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Axially Symmetric Wave Propagation in a Two-Layered Cylinder

OCTOBER 1965

Prepared by

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Aerodynamics and Propulsion Research Laboratory

Prepared for BALLISTIC SYSTEMS AND SPACE SYSTEMS DIVISIONS

AIR FORCE SYSTEMS COMMAND

LOS ANGELES AIR FORCE STATION

Los Angeles, California



LABORATORY OPERATIONS •

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Report No.
TDR-469(5240-10)-18

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TWO-LAYERED CYLINDER**

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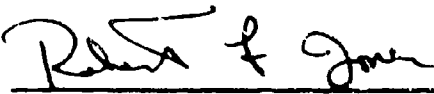
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This technical documentary report has been reviewed and is approved for publication and dissemination. The conclusions and findings contained herein do not necessarily represent an official Air Force position.

For Space Systems Division
Air Force Systems Command


Robert F. Jones
Captain, USAF

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ABSTRACT

The linear theory of elasticity is used to investigate axially symmetric wave propagation in an infinitely long two-layered cylinder. Each material is taken to be homogeneous and isotropic. A perfect bond is assumed at the interface, while the inner and outer boundaries of the composite cylinder are treated as traction-free. The dispersion determinant relating phase velocity and wave number for a harmonic train of waves satisfying these boundary conditions is presented. The character of the dispersion equation is investigated analytically and numerically. Stress and displacement distributions are also presented for the numerical example. Comparisons are made with an approximate solution of the same problem obtained by means of a thin shell theory incorporating thickness-shear deformation of each layer.

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I. INTRODUCTION

In recent years considerable attention has been focused on multi-layered shells. Often a sandwich-type construction is used to lighten the weight of a shell structure; in other instances, a protective layer is bonded to a shell as, for example, in the case of a re-entry vehicle heat shield or a rocket nozzle liner. Two-layered thick shell configurations are also used in solid propellant rocket motors.

This study employs the linear theory of elasticity and treats the propagation of a train of waves in an infinitely long, two-layered cylinder; each layer is homogeneous and isotropic. Computations based on this solution are used in an assessment of the accuracy of an approximate two-layered shell theory that was presented recently.¹ This latter theory included the effects of shear deformation and rotatory inertia.* In Ref. 1, a partial comparison is made between the shell theory and the solutions obtained in the present work using the linear theory of elasticity. Dispersion curves were compared in detail, and a few displacement distribution comparisons were made. The present work gives detailed displacement comparisons over a larger frequency and wave number regime and also presents stress distribution comparisons.

The propagation of waves in cylindrically bounded media has been extensively investigated. Although most of the work has been limited to cylinders of a single material, it is of interest to recall a few of the more pertinent references. Pochhammer² and Chree³ first formulated the problem for solid cylindrical bars. Ghosh⁴ formulated the problem for hollow cylindrical bars but presented no calculations. Later, Gazis^{5,6,7} and Greenspan^{8,9} made extensive numerical calculations for the vibrations of a

* For a more extensive bibliography dealing with the thin shell literature of layered shells see Ref. 1.

hollow cylinder and compared them with several approximate shell theories such as those of Herrmann and Mirsky^{10, 11, 12} and Naghdi and Cooper.^{13, 14} Other investigators include Bird¹⁵ and Bird, Hart, and McClure.¹⁶

The vibrations of a multilayered cylinder using the equations of the linear theory of elasticity have not been extensively investigated. Baltrukonis, Gottenberg, and Schreiner¹⁷ treated simple thickness-shear vibrations of a two-layered cylinder, and McNiven, Sackman, and Shah¹⁸ treated propagation of axially symmetric waves in solid bars with an outer finite layer. Other than these, the authors know of no other references treating multilayered cylinders by the linear theory of elasticity.

Layered half-planes have been extensively treated by geophysicists,¹⁹ but their work is not particularly of interest here since geophysical earth models always have one infinite layer. Of the plane two medium problems with finite layers, the symmetrical sandwich two-dimensional beam is treated by Saito and Sato,²⁰ and the asymmetrical two-layered counterpart is treated by Jones.²¹ The last reference is especially apropos since it is shown here that the wave propagation solution of the two-layered cylinder problem degenerates into the solution of the plane two layered medium problem when the wavelength becomes sufficiently small compared to the thickness.

The present analysis is formulated in terms of displacement potentials. A solution in the form of an infinite train of axially symmetric waves is assumed. To satisfy boundary conditions the phase velocity (or frequency) must depend on wave number in such a way that an eighth-order determinant vanishes. Due to its complexity little analytical progress can be made with this dispersion determinant except in special cases. However, for infinitely long waves the determinant reduces to a product of two fourth-order determinants whose frequency roots correspond to vibrations with either purely axial or purely radial motion. Alternatively, for very short waves the determinant reduces to a form given previously by Jones²¹ for a plane two-layered medium. Here the

phase velocity roots correspond to Rayleigh waves on the free surfaces and a possible Stoneley wave at the interface.* For waves of intermediate length, roots of the dispersion determinant are found numerically with a digital computer program described herein. Displacement and stress distributions corresponding to these roots are also found for a specific numerical example, and these are used to estimate the accuracy of the previously mentioned shell theory.

It is concluded that a Timoshenko-type shell theory gives good agreement with the present exact solution in a region of applicability encompassing low enough frequencies and large enough wavelengths. It appears that to extend this region of applicability one must use a shell theory incorporating thickness-stretch motion. On the basis of the displacements obtained from the exact theory, a linear distribution of radial motion does not appear to be an unreasonable first approximation for a thickness-stretch theory.

* A Stoneley wave may or may not exist at an interface depending on the elastic properties and densities of the two media (Ref. 19, p. 113).

II. ANALYSIS

Consider a doubly infinite hollow cylinder composed of two homogeneous, isotropic, elastic media with Lamé constants λ_1 , μ_1 , λ_2 , and μ_2 and densities ρ_1 and ρ_2 , where subscript 1 refers to the outer layer and subscript 2 refers to the inner layer. The cylinders are perfectly bonded together at the interface. Cylindrical coordinates r , θ , and z are employed. The interface radius is denoted by a , and the thicknesses are h_1 and h_2 .

Written in terms of the potential functions ϕ and ψ the equations of elasticity for motions with torsionless axial symmetry are

$$\nabla^2 \phi = \frac{1}{\lambda} \phi_{,tt} \quad , \quad \nabla^2 \psi = \frac{1}{\mu} \psi_{,tt} \quad (1)$$

where the displacements and stresses may be generated from the potential functions by

$$u_\theta = 0 \quad , \quad u_r = \phi_{,r} + \psi_{,rz} \quad , \quad w = \phi_{,z} - \frac{1}{r} (r\psi_{,r})_{,r} \quad (2)$$

$$\sigma_{rr} = \lambda \left(\frac{1}{r} \phi_{,r} + \phi_{,rr} + \phi_{,zz} \right) + 2\mu (\phi_{,r} + \psi_{,rz})_{,r} \quad (3)$$

$$\sigma_{rz} = \mu \left(2\phi_{,rz} + \psi_{,zz} - \psi_{,rr} - \frac{1}{r} \psi_{,r} \right)_{,r} \quad (4)$$

$$\sigma_{zz} = \lambda \left(\frac{1}{r} \phi_{,r} + \phi_{,rr} + \phi_{,zz} \right) + 2\mu \left[\phi_{,zz} - \frac{1}{r} (r\psi_{,r})_{,r} \right] \quad (5)$$

The terms u and w are the radial and axial displacements, respectively, and σ_{rr} , σ_{rz} , and σ_{zz} are the stress components. The expression for $\sigma_{\theta\theta}$ is omitted since it is not essential to the problem. All other stress components are zero and α and β are the dilatational and equivoluminal wave speeds defined by

$$\alpha^2 = \frac{\lambda + 2\mu}{\rho}, \quad \beta^2 = \frac{\mu}{\rho} \quad (6)$$

Equations (1)-(5) hold in either layer provided appropriate values of α , β , λ and μ are used.

If a Rayleigh train of waves is assumed, then all quantities are considered to vary as

$$w = \bar{w}(r)e^{i(kz - \omega t)}, \quad \sigma_{rr} = \bar{\sigma}_{rr}(r)e^{i(kz - \omega t)} \quad \text{etc.} \quad (7)$$

and the solution to Eqs. (1) is (the factor $e^{i(kz - \omega t)}$ has been suppressed for compactness)

$$\phi = AI_0(kr) + BK_0(kr) \quad (8)$$

$$ik\psi = CI_0(kr) + DK_0(kr)$$

Here I_0 and K_0 are modified Bessel functions of the first and second kind and zeroth order. The solutions Eq. (8) hold in either medium provided the parameters appropriate to each medium are used in the equations. Thus there will be two sets of ϵ 's and δ 's:

$$\epsilon_1^2 = 1 - \frac{c^2}{\alpha_1^2}, \quad \epsilon_2^2 = 1 - \frac{c^2}{\alpha_2^2} \quad (9)$$

(cont.)

$$\delta_1^2 = 1 - \frac{c^2}{\beta_1^2}, \quad \delta_2^2 = 1 - \frac{c^2}{\beta_2^2} \quad (9)$$

where

$$c^2 = \frac{\omega^2}{k^2} \quad (10)$$

Also there will be two sets of constants A_1, A_2 and B_1, B_2 , etc., to be determined by application of boundary conditions on $\sigma_{rr}, \sigma_{rz}, u_r$ and w . Expressing the displacements and stresses in terms of the solutions, Eq. (8), one obtains

$$\begin{aligned} \frac{\sigma_{rr}}{\mu} = k^2 & \left\{ \left[A(1 + \delta^2)I_0(k\epsilon r) - 2\epsilon^2 \frac{I_1(k\epsilon r)}{k\epsilon r} \right] + B \left[(1 + \delta^2)K_0(k\epsilon r) + 2\epsilon^2 \frac{K_1(k\epsilon r)}{k\epsilon r} \right] \right\} \\ & + 2k^2\delta^2 \left\{ C \left[I_0(k\delta r) - \frac{I_1(k\delta r)}{k\delta r} \right] + D \left[K_0(k\delta r) + \frac{K_1(k\delta r)}{k\delta r} \right] \right\} \end{aligned} \quad (11)$$

$$\frac{\sigma_{rz}}{\mu} = 2ik^2\epsilon \left[AI_1(k\epsilon r) - BK_1(k\epsilon r) \right] + ik^2\delta \left[C(1 + \delta^2)I_1(k\delta r) - D(1 + \delta^2)K_1(k\delta r) \right] \quad (12)$$

$$u_r = k \left[A\epsilon I_1(k\epsilon r) - B\epsilon K_1(k\epsilon r) + C\delta I_1(k\delta r) - D\delta K_1(k\delta r) \right] \quad (13)$$

$$w = ik \left[AI_0(k\epsilon r) + BK_0(k\epsilon r) + C\delta^2 I_0(k\delta r) + D\delta^2 K_0(k\delta r) \right] \quad (14)$$

The above expressions are valid in each region provided appropriate values of $A_i, B_i, C_i, D_i, \epsilon_i, \delta_i, \mu_i$ ($i = 1, 2$) are used.

The boundary conditions for the free vibration problem are

$$\left. \begin{aligned}
 &\sigma_{rr}^{(1)} - \sigma_{rz}^{(1)} = 0 \quad \text{at} \quad r = a + h_1 \\
 &\sigma_{rr}^{(2)} - \sigma_{rz}^{(2)} = 0 \quad \text{at} \quad r = a - h_2 \\
 &\text{and} \\
 &\sigma_{rr}^{(1)} = \sigma_{rr}^{(2)}, \quad \sigma_{rz}^{(1)} = \sigma_{rz}^{(2)}, \quad u^{(1)} = u^{(2)} \\
 &w^{(1)} = w^{(2)} \quad \text{at} \quad r = a
 \end{aligned} \right\} \quad (15)$$

The superscripts indicate quantities in medium 1 or 2. From the boundary condition, Eq. (15), one obtains with the aid of Eqs. (11)-(14) eight linear homogeneous equations for the eight constants $A_1, \dots, D_2$. Since the equations are long and their formulations are straightforward, the equations are presented in the Appendix rather than in the text.

To assure nontrivial values of $A_1, B_1, \dots, C_2, D_2$, the determinant of their coefficients must be set equal to zero. This constitutes the dispersion equation. The determinant is as follows:

P_{11}	Q_{11}	R_{11}	S_{11}	0	0	0
T_{11}	$-U_{11}$	V_{11}	$-W_{11}$	0	0	0
$\mu_1 P_{10}$	$\mu_1 Q_{10}$	$\mu_1 R_{10}$	$\mu_1 S_{10}$	$-\mu_2 P_{20}$	$-\mu_2 Q_{20}$	$-\mu_2 S_{20}$
$\mu_1 T_{10}$	$-\mu_1 U_{10}$	$\mu_1 V_{10}$	$-\mu_1 W_{10}$	$-\mu_2 T_{20}$	$-\mu_2 U_{20}$	$-\mu_2 V_{20}$
$\delta_{11}^2 I_1(k_1, a)$	$-\delta_{11}^2 K_1(k_1, a)$	$\delta_{11}^2 I_1(k_1, a)$	$-\delta_{11}^2 K_1(k_1, a)$	$-\delta_{21}^2 I_1(k_2, a)$	$-\delta_{21}^2 K_1(k_2, a)$	$\delta_{21}^2 K_1(k_2, a)$
$I_0(k_1, a)$	$K_0(k_1, a)$	$\delta_{11}^2 I_0(k_1, a)$	$\delta_{11}^2 K_0(k_1, a)$	$-\delta_{21}^2 I_0(k_2, a)$	$-\delta_{21}^2 K_0(k_2, a)$	$-\delta_{21}^2 K_0(k_2, a)$
0	0	0	0	$-P_{22}$	$-Q_{22}$	$-S_{22}$
0	0	0	0	$-T_{22}$	U_{22}	W_{22}

(16)

where, for brevity, the quantities $P_{11}, \dots, W_{22}$ are expressions containing Bessel functions and are defined in the Appendix.

The determinant, Eq. (16), is so complicated that little can be done to obtain any general analytical results. However, simpler forms of the determinant are obtainable for the limiting cases of very long and very short waves.

For infinite wavelengths, i. e., $k = 0$, the solutions, Eq. (8), become independent of z , and the determinant degenerates to the product of two fourth-order determinants. A more straightforward derivation of these fourth-order determinants involves repeating the calculations with the z dependence excluded from the outset. Then it is clear that one of the fourth-order determinants corresponds to purely radial motions while the other corresponds to purely axial motions.

Equating the radial motion determinant to zero one obtains the frequency equation for simple thickness stretch vibrations:

[illegible]

where J_0 , M_0 , J_1 , M_1 are Bessel functions of the first and second kinds of orders zero and one, respectively.

Equating the axial motion determinant to zero one obtains the frequency equation for axial shear vibrations

$$\begin{vmatrix} J_1\left[\frac{\omega}{\beta_1}(z+h_1)\right] & N_1\left[\frac{\omega}{\beta_1}(z+h_1)\right] & 0 & 0 \\ J_1\left(\frac{\omega}{\beta_1}a\right) & N_1\left(\frac{\omega}{\beta_1}a\right) & -\frac{\mu_2\beta_1}{\mu_1\beta_2}J_1\left(\frac{\omega}{\beta_2}a\right) & -\frac{\mu_2\beta_1}{\mu_1\beta_2}N_1\left(\frac{\omega}{\beta_2}a\right) \\ J_0\left(\frac{\omega}{\beta_1}a\right) & N_0\left(\frac{\omega}{\beta_1}a\right) & -J_0\left(\frac{\omega}{\beta_2}a\right) & -N_0\left(\frac{\omega}{\beta_2}a\right) \\ 0 & 0 & J_1\left[\frac{\omega}{\beta_2}(a-h_2)\right] & N_1\left[\frac{\omega}{\beta_2}(a-h_2)\right] \end{vmatrix} = 0 \quad (18)$$

Equation (18) has been obtained by Baltrukonis, et al.¹⁷

For very short wavelengths compared to the interface radius of the cylinder, it would be expected that propagation of axially symmetric waves in a two-layered cylinder would differ very little from propagation of straight crested waves in a plane two-layered medium at least for a thin cylinder. The truth of this supposition can be shown analytically in Eq. (16) by replacing I_0 , I_1 , K_0 , and K_1 by their asymptotic values for large arguments:

$$\begin{aligned} I_0, I_1(x) &\approx \frac{e^x}{(2\pi x)^{1/2}} \\ K_0, K_1(x) &= \left(\frac{\pi}{2x}\right)^{1/2} e^{-x} \end{aligned} \quad (19)$$

($x \gg 1$)

With these substitutions and a slight redefinition of coefficients in Eqs. (A-1)-(A-8), the dispersion determinant becomes the same as that given by Jones²¹ for the plane two-layered problem. As would be expected, the waves degenerate into two Rayleigh surface waves, one in each medium, and a possible Stoneley wave at the interface.

For wavelengths of intermediate size it is difficult to extract much information about the character of the possible wave propagation solutions except by considering specific numerical examples. This is done in the following section where the results of the numerical example are also used to check the accuracy of the shell theory given in Ref. 1.

III. COMPUTATIONS

In order to investigate the character of the wave propagation solutions governed by Eqs. (A-1)-(A-8) numerical analysis was employed. A program for the IBM 7094 computer was developed at the Aerospace Corporation for determining phase velocities (and from them, frequencies) satisfying Eq. (16). To facilitate the use of the present solution for assessing the range of validity of approximate shell theories, displacement and stress distributions through the cylinder thickness are calculated as well.

For a given numerical problem valid computer results are obtained only for a limited range of wavelengths. The largest number available for routine calculation is $10^{38} \sim \exp(88)$, while for large wave numbers the I 's (modified Bessel functions of the first kind) are of the order of $\exp[(kH)(a/H)]$. Here $H = h_1 + h_2$ is the total wall thickness of the cylinder. Thus, for valid computer results, kH cannot be much larger than $88(H/a)$. In the example to be considered here a/H is 30 so that $kH < 3$ gives a fair estimate of the range amenable to computations based on Eq. (16). For larger values of kH the wavelength is short compared to the radius of the cylinder, the asymptotic expressions of Eq. (19) are appropriate, and satisfactory numerical results are obtained using a computer program based on the plane medium equations of Ref. 21.

Properties of the cylinder chosen for detailed numerical study are given in Table I. This example was used to check the validity of the approximate shell theory of Ref. 1 where dispersion curves were compared for low frequencies and large wavelengths. Additional dispersion curves are presented here for the first nine modes of axially symmetric wave propagation. Distributions of displacements and stresses through the thickness of the cylinder are also presented, and for the first four modes these are compared with the distributions predicted by the approximate shell theory.¹

Table 1. Numerical Properties of the Cylinder Studied

Properties	Outer Layer	Inner Layer
Material Constants	$E_1 = 4.00 \times 10^6 \text{ lb/in.}^2$	$E_2 = 30.0 \times 10^6 \text{ lb/in.}^2$
	$\nu_1 = 1/4$	$\nu_2 = 1/3$
	$\rho_1 = \left(\frac{0.080}{386}\right) \text{ lb-sec}^2/\text{in.}^4$	$\rho_2 = \left(\frac{0.286}{386}\right) \text{ lb-sec}^2/\text{in.}^4$

Geometrical Parameters

$$\frac{a}{H} = 30 \quad \frac{h_1}{H} = 0.3 \quad \frac{h_2}{H} = 0.7 \quad (H = h_1 + h_2)$$

A. DISPERSION CURVES

Dispersion curves for the cylinder of Table 1 appear in Figs. 1 and 2. In both graphs, the abscissa is nondimensional wave number $\Delta = kH$. In Fig. 1 the ordinate is nondimensional phase velocity $s = c/\beta^* = \omega/kB^*$, and in Fig. 2 it is nondimensional frequency $\Omega = \omega H/\beta^*$. The reference velocity $\beta^* = (\beta_1 + \beta_2)/2$.

The approach of the phase velocity to various limiting values may be examined in Fig. 1. For instance, for very long waves, the phase velocity of the first mode approximates that of "bar" waves. Also, for very short waves the phase velocity of the first mode approaches the Rayleigh wave speed of the slower medium. For the second and higher modes the phase velocity becomes very large as the wave number is made smaller. Therefore, the long wavelength behavior of these modes is better examined in the frequency wave number plot of Fig. 2.

In Fig. 2 it is observed that curves for all the modes except the first have a finite frequency intercept for zero wave number. These frequencies and the character of the motion for each of the modes are summarized in Table II. Other features of the curves presented in Fig. 2 include the relative minimum

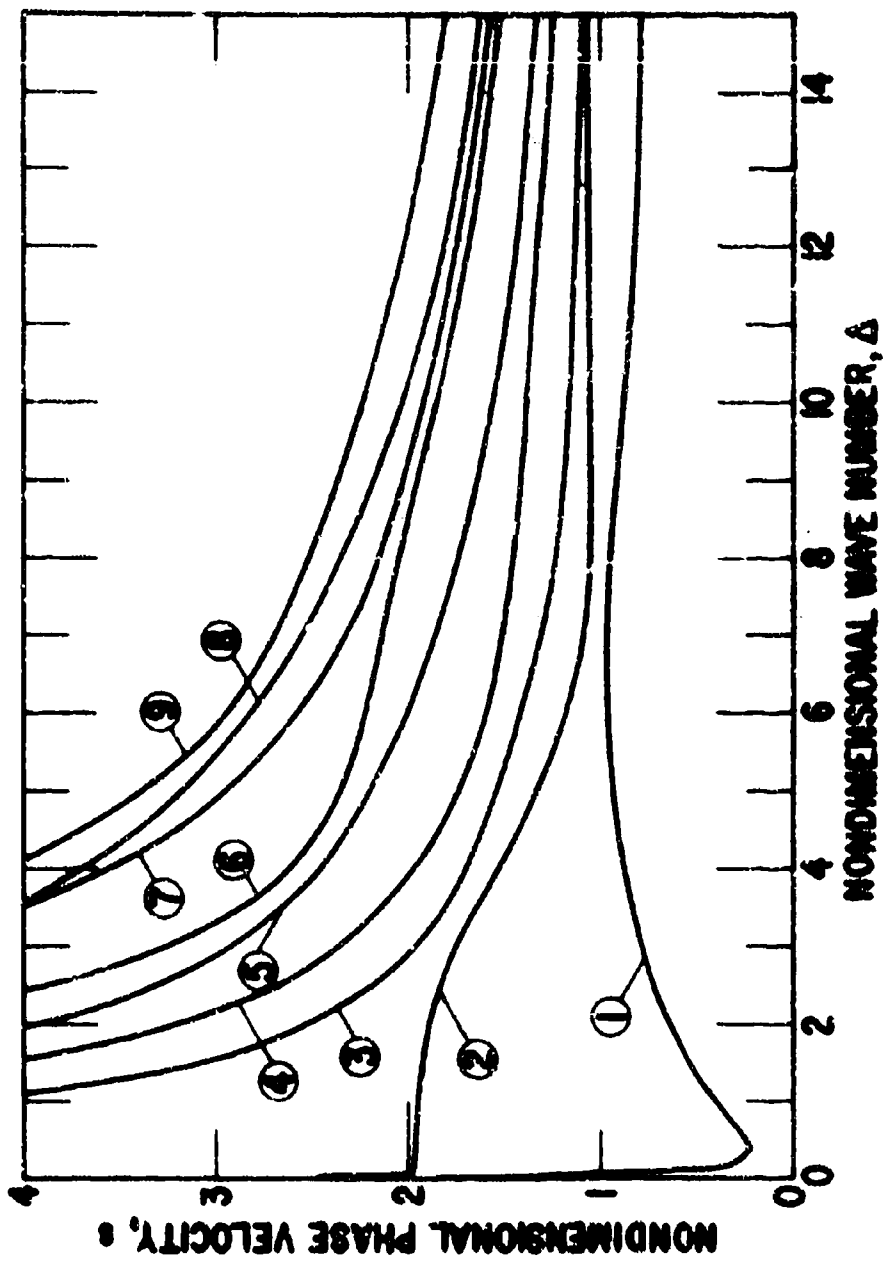


Fig. 1. Phase Velocity vs Wave Number

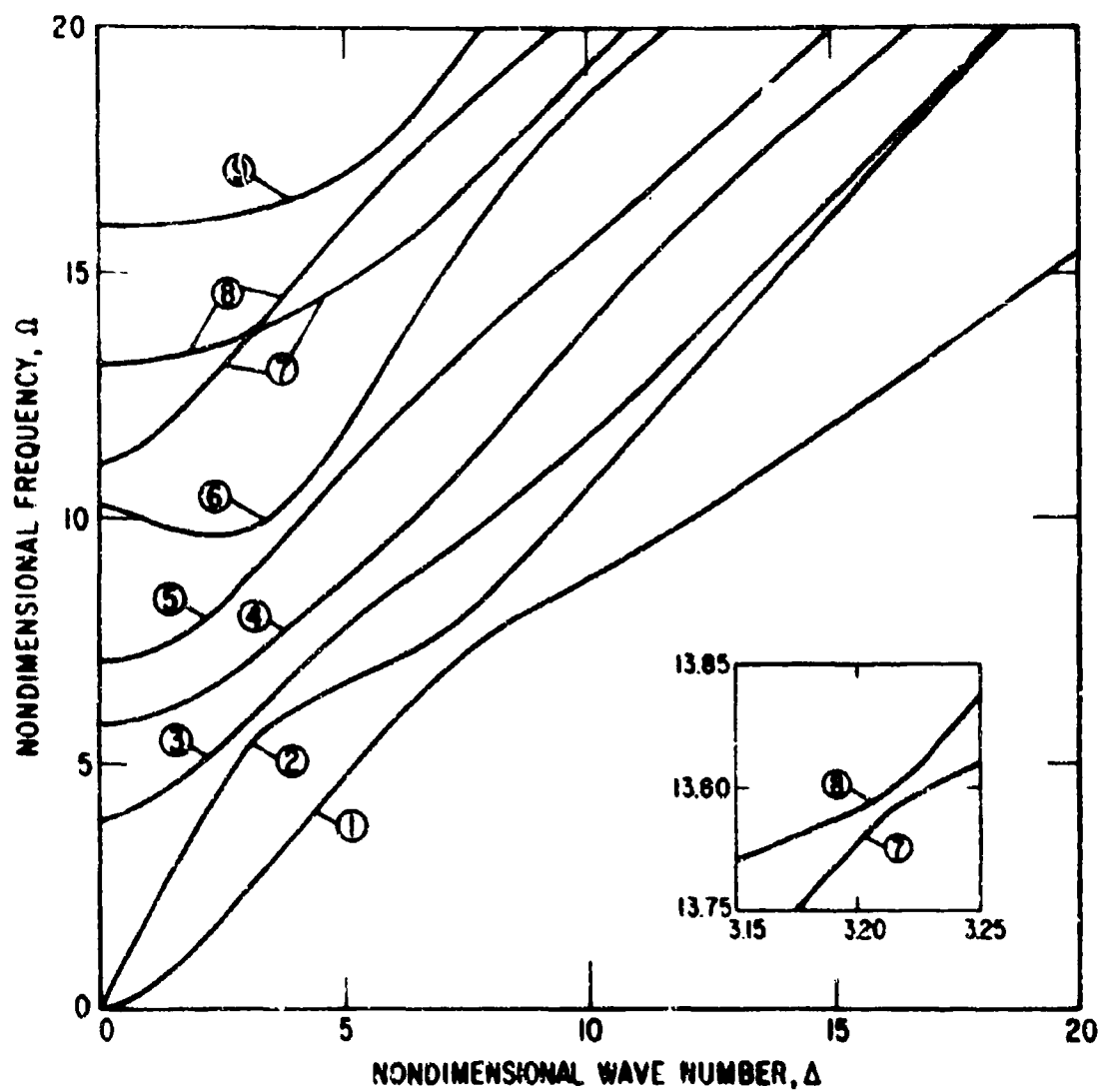


Fig. 2. Frequency vs Wave Number

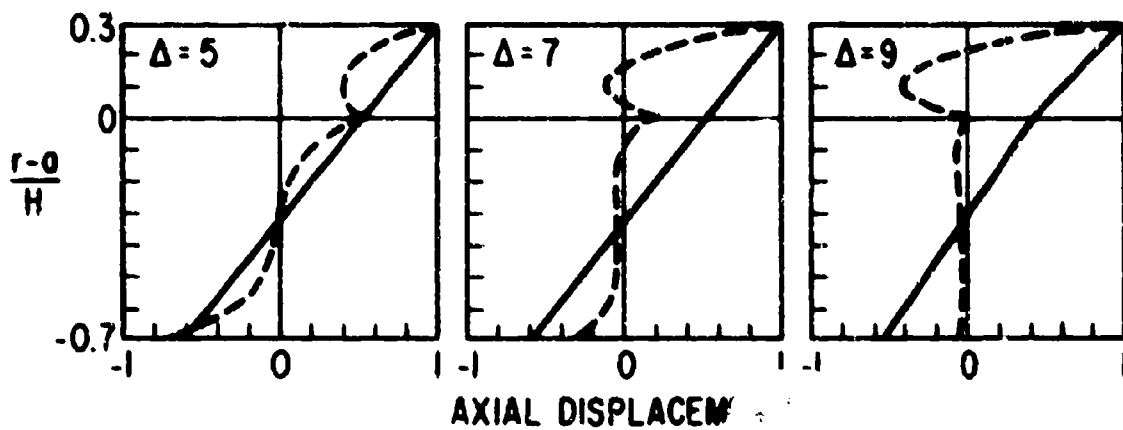
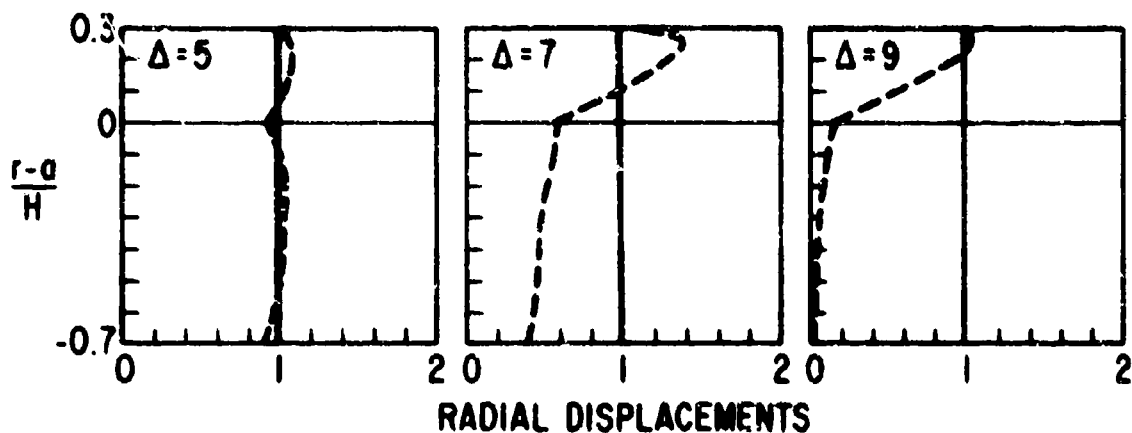
Table II. Character of the Infinite Wavelength Axially Symmetric Vibrations of the Two-Layered Cylinder of Table I

Mode	Ω Frequency	Character of Motion
1	0	zero nodes, axial motion
2	0.066	zero nodes, radial motion, "ring" vibration
3	3.86	1 node, axial motion, thickness shear
4	5.80	2 nodes, axial motion, thickness shear
5	7.07	1 node, radial motion, thickness stretch
6	10.2	3 nodes, axial motion, thickness shear
7	11.1	2 nodes, radial motion, thickness stretch
8	13.1	4 nodes, axial motion, thickness shear
9	15.9	5 nodes, axial motion, thickness shear

of mode 6 near $\Delta = 2.5$ (this corresponds to zero group velocity) and the close approach of modes 7 and 8 near $\Delta = 3.1$. The inset of Fig. 2 shows this close approach on a magnified scale.

B. DISPLACEMENTS

In Ref. 1 an approximate shell theory is developed and partially compared with the present exact solutions. Figures 3 through 7 extend the comparison of displacements over the entire range of interest. Due to the eigenvalue nature of the problem considered the absolute magnitudes of the displacements are undetermined since multiplication by a constant (normalisation) factor is permissible. Therefore, our comparisons are of the shapes of the displacement distributions, the magnitude having been adjusted to make the present exact theory and the shell theory of Ref. 1 agree at some convenient value of r . Solution for both theories of a forced motion problem would permit comparisons of magnitudes as well. However, examination of the shapes alone of the displacement distributions is of interest since formulations of higher order shell theories generally postulate functional forms for the dependence of the displacements on the thickness coordinate.



— SHELL THEORY - - - - ELASTICITY THEORY

Fig. 3. First Mode Displacement Distribution

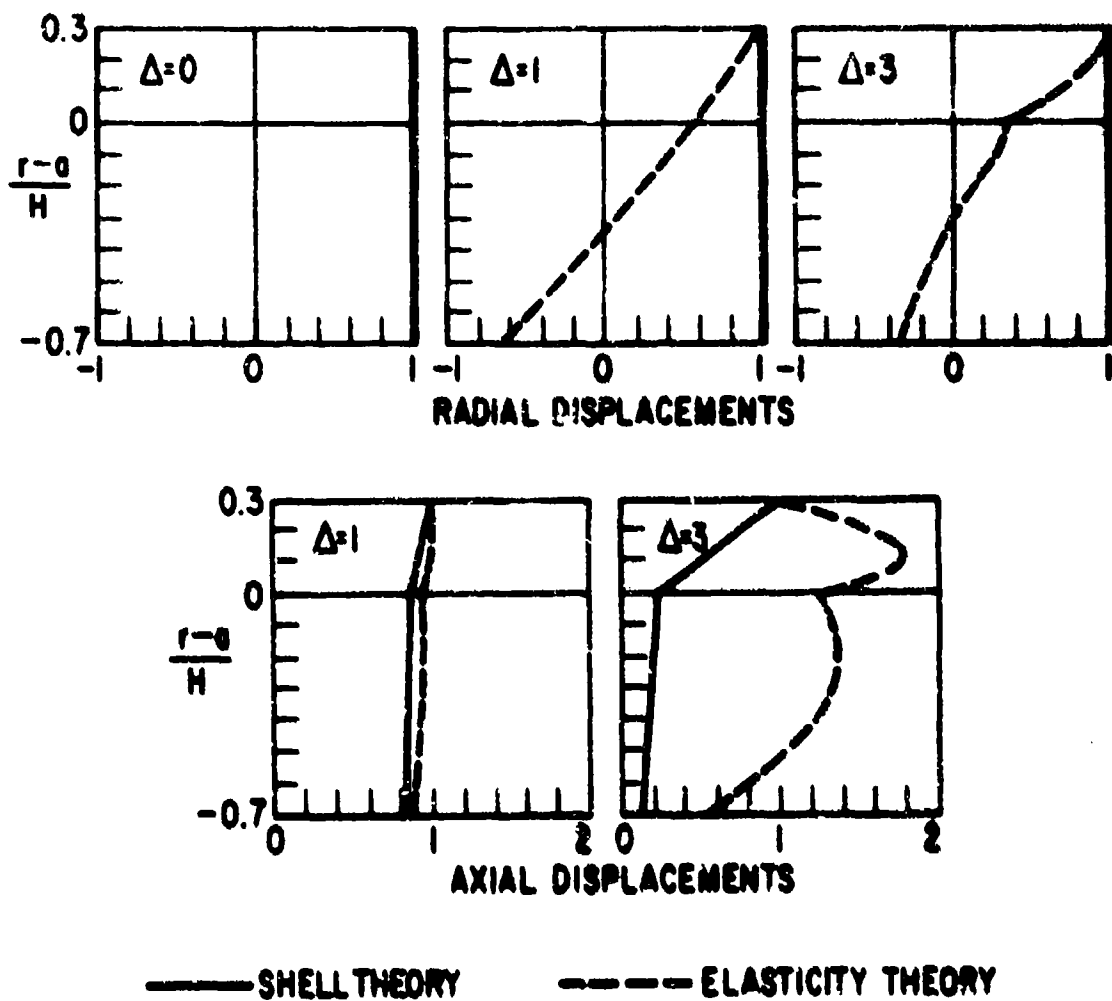


Fig. 4. Second Mode Displacement Distributions

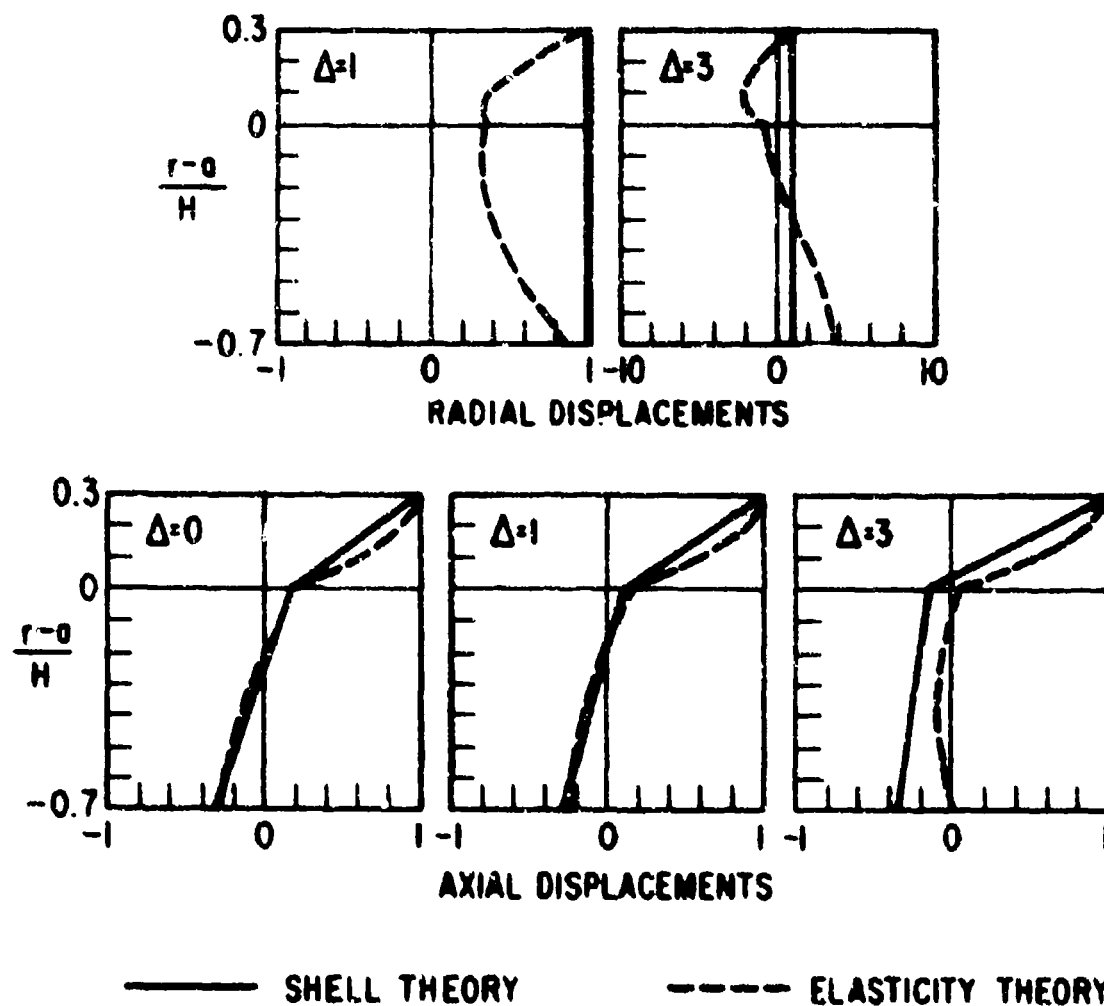


Fig. 5. Third Mode Displacement Distributions

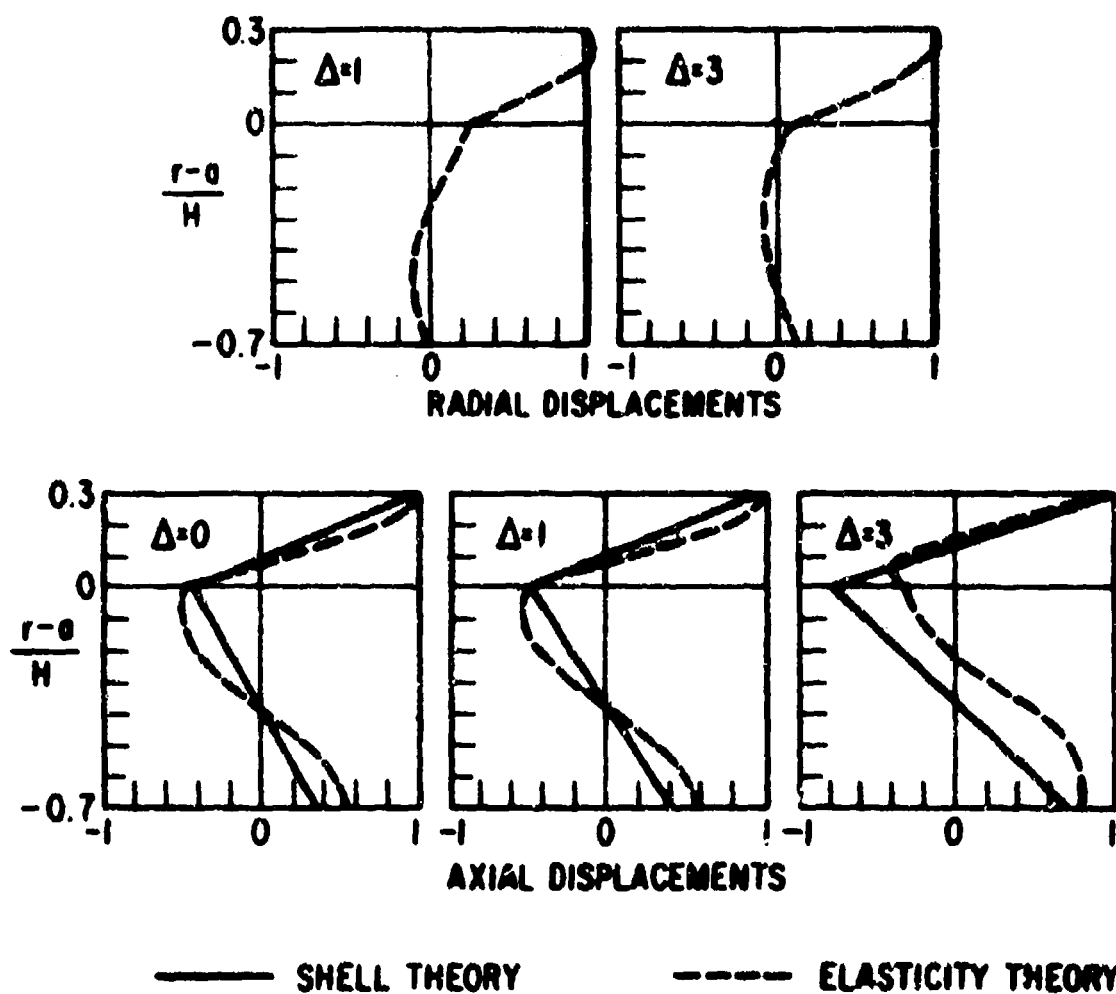


Fig. 6. Fourth Mode Displacement Distributions

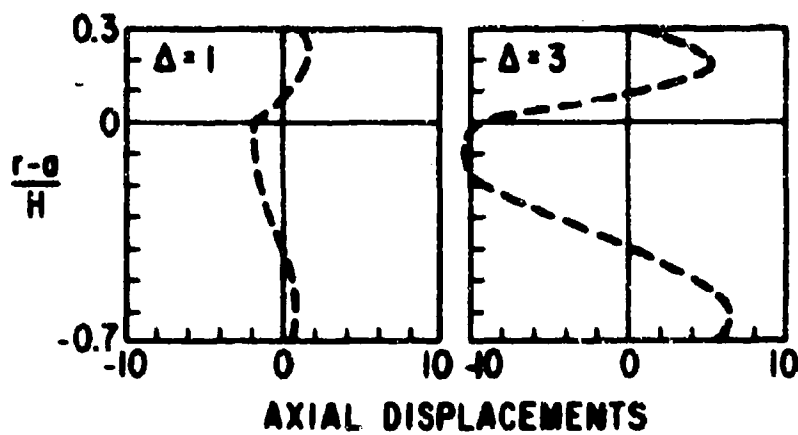
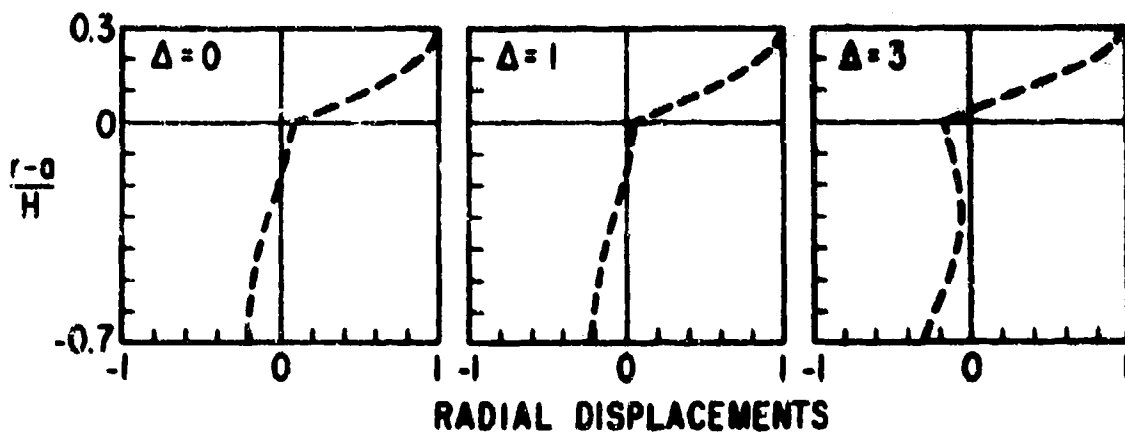


Fig. 7. Fifth Mode Displacement Distributions

For the present comparisons the axial displacements have been normalized to make exact theory and shell theory displacements equal to one at the outer surface. This normalization, of course, fixes definite ratios between the radial displacements of the two theories at every location in the shell (except in the case of infinite wavelength where axial and radial motion are uncoupled). The radial displacements need nowhere be equal. Rather than present them in such a form we have taken the liberty of introducing a different normalization of the radial displacements. The radial displacements from both theories are made equal to one at the outer surface. Such a normalization is necessary for uncoupled radial motion, and moreover it permits the use of a uniform scale for nearly all the radial displacement plots. The information suppressed by this normalization convention, that is, the ratio of the maximum outer surface radial displacement to maximum outer surface axial displacement, is presented in Table III.

In Ref. 1 the first mode axial and radial displacements (as well as the dispersion curves) were compared for a limited range of $\Delta < 3.0$. Figure 3 extends this comparison up to $\Delta = 9.0$. Since several curves for $0 < \Delta \leq 3.0$ were presented in Ref. 1, the curves in Fig. 3 start at $\Delta = 5.0$. Further, for $\Delta < 3.0$, there is little variation of radial displacement through the thickness, and the axial displacement differs little from the bi-linear distribution assumed in the shell theory. Starting at $\Delta = 5.0$ the nonuniformity of radial displacement begins to become pronounced, and the axial displacement becomes increasingly nonlinear.

Previous analyses^{1, 21} show that as the wave number becomes increasingly large the first mode decays from a flexural mode into a Rayleigh wave in the slower medium. It is apparent in the curves for $\Delta = 9.0$ that this transition is nearly complete: the motion is concentrated almost entirely in the outer medium. One might expect certain of the displacement plots to exhibit other surface or interface waves. However, with the parameters used for these two media, calculations show that Stoneley waves do not exist at the interface (Ref. 19, p. 113). Theory predicts that for sufficiently high wave numbers a Rayleigh wave will form in the faster (inner) medium. Such a wave

Table III. Ratios of Outer Surface Maximum Radial Displacement to Outer Surface Maximum Axial Displacement

Mode	Wave Number, Δ	Displacement Ratio	
		Exact Theory	Shell Theory
1	5	1.089	1.370
	7	1.460	1.531
	9	1.630	1.739
2	0	"	"
	1	-0.277	-0.0165
	3	-2.621	-0.045
3	0	0	0
	1	-0.1025	-0.0467
	3	-0.0428	-0.0521
4	0	0	0
	1	0.522	0.0277
	3	1.886	0.0962
5	0	"	--
	1	-11.96	--
	3	-21.50	--

is manifested in a more complicated way called "terracing,"²² and its consideration would require investigation of a larger range of Δ than has been undertaken for the present work.

Figure 4 shows the second mode displacements for $\Delta = 0, 1$, and 3. For $\Delta = 0$, the motion is totally radial, and no plot of axial displacements is necessary. In contrast to the first mode, the displacement curves degenerate from the shell theory quite rapidly with a phase reversal in the radial displacement already apparent at $\Delta = 1.0$. By $\Delta = 3.0$ the nonlinearity of the curves is pronounced. They are totally different from those of the shell theory. The

radial displacement curves for both $\Delta = 1.0$ and 3.0 are evidence that considerable thickness-stretch deformation is present.

Figures 5 and 6 show the displacements for the third and fourth modes. They are predominantly thickness-shear modes. At $\Delta = 0$ the motion is purely axial; therefore no radial displacement plots appear. In both figures, the predictions from shell theory compare quite well with the exact solution, at least for the axial displacements. As in the second mode there is sufficient thickness-stretch motion present to cause the appearance of nodes in most of the radial displacement plots. Note in Fig. 5 that the radial displacements for $\Delta = 3$ are plotted to a different scale than the rest.

Figure 7 shows the displacements for the fifth mode, the first mode not predicted by shell theory. At $\Delta = 0$ the motion is purely radial or thickness-stretch. At least for small wave number the shapes of the radial displacement curves are such that a linear distribution might offer a good approximation. Thus, a higher order shell theory admitting a linear distribution of radial displacements might give fair results for the long wavelength portion of the fifth mode. Such a shell theory has been presented by Mirsky²³ for a single-layer cylindrical shell. Resuming the examination of Figure 7 one sees that for finite wave numbers axial motion is present in a form that resembles double-node thickness shear. However, the axial motion shows little tendency toward linearity, indicating that (the same as in the lower modes) a shell theory with linear axial displacement variation through the thickness will probably not give a reasonable approximation for larger wave numbers.

In summary the comparisons of the displacements for the four lowest modes shows that the second mode yields the poorest comparison between exact and shell theory. Although this result is surprising, it could have been anticipated since the second mode dispersion curve deviates more from that predicted by shell theory for lower values of Δ than for the other modes. It would appear that the incorporation of thickness-stretch deformations is the next essential step in improving the shell theory for high frequency, short wavelength use.

C. STRESSES

Figures 8 through 11 describe the axial stress (σ_{xx}) and the shear stress (τ_{xz}) distributions. As with the displacements the absolute magnitude of the stresses cannot be determined. The stresses from both the exact and shell theories were normalized so that the greatest axial stress at the outer edges or at the interface was set equal to either ± 1.0 . This also fixes the magnitudes of the shear stresses.

Figures 8 and 9 show, respectively, the first mode axial and shear stresses. It is seen in Fig. 8 that for $\Delta \leq 3$ there is good agreement with the shell theory. However, as Δ increases the agreement deteriorates. By $\Delta = 7$ there is only fair agreement, and by $\Delta = 9$ there is almost none. The latter point is confirmed by the previous observation that by $\Delta = 9$ there is a Rayleigh wave formed in the outer medium. Figure 9 is probably of more interest since an expected mode of failure of multilayered shells is in shear failure of the bond. The shear stress distribution follows roughly the same pattern as the axial stresses, being close to the shell theory for $\Delta \leq 5$ and being totally different by $\Delta = 9$.

Figure 10 shows the second mode stresses. As might be expected there is not as good an agreement as with the first mode. By $\Delta = 3.0$ the shell theory stress distributions are beginning to diverge sharply from the exact ones.

Figure 11 shows third mode stress distributions. There is good agreement with shell theory only for $\Delta \leq 1$. By $\Delta = 2$ there is considerable divergence in the stresses, and by $\Delta = 3$ the exact theory curves bear little resemblance in shape to those from the shell theory. The fourth mode curves, while not presented, show a pattern similar to the third mode. It is worth noting that although the stress comparison plots are useful for illustrating qualitative differences between the theories the quantitative interpretation can be quite sensitive to the particular normalization convention that is adopted.

In conclusion it is felt that if any shell theory is to be improved by the introduction of more dependent variables, thickness-stretch motion is essential. For still higher modes to yield agreement it would be necessary to add additional shear deformations corresponding to warping of initially plane cross sections.

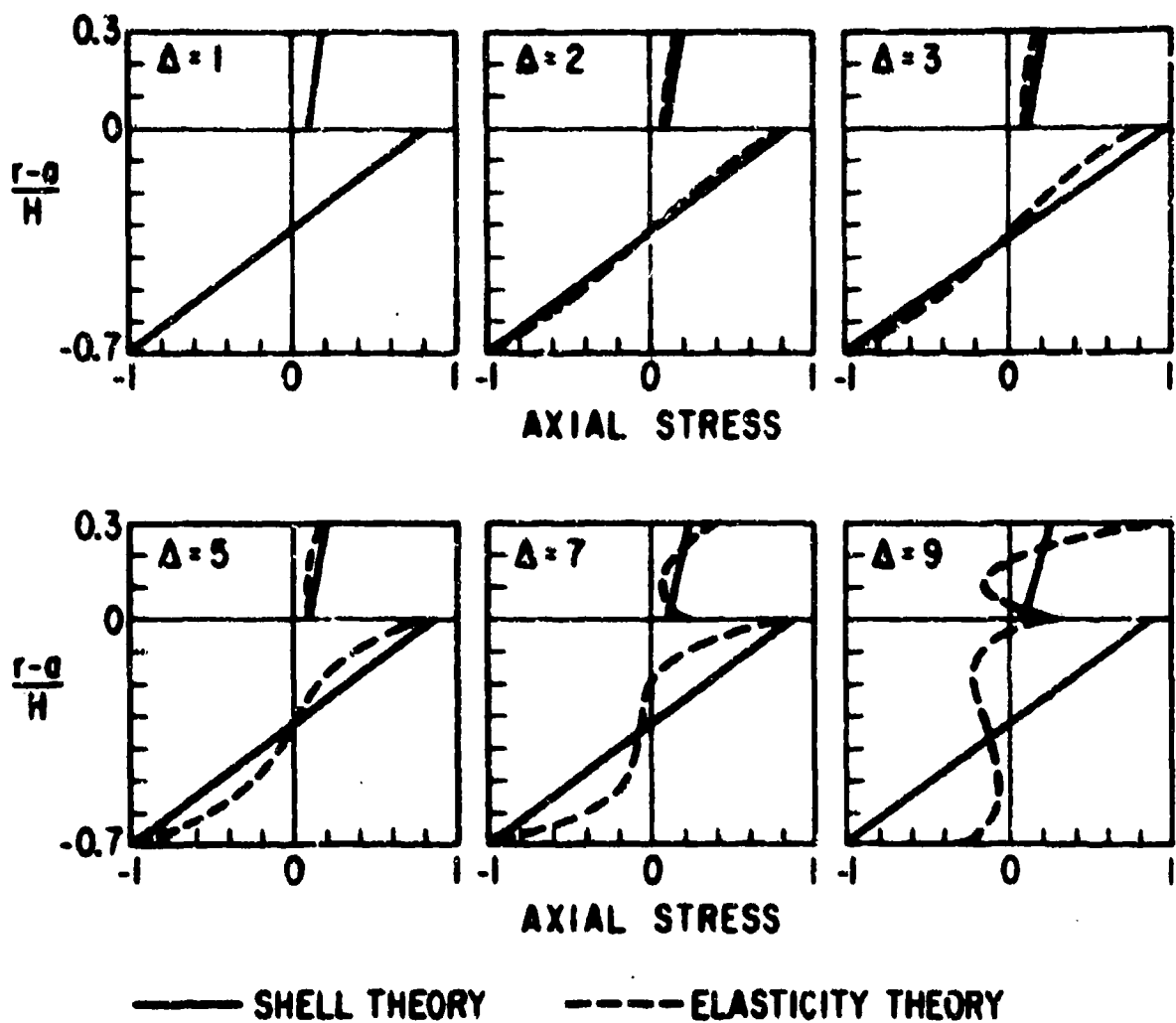


Fig. 8. First Mode Axial Stress Distributions

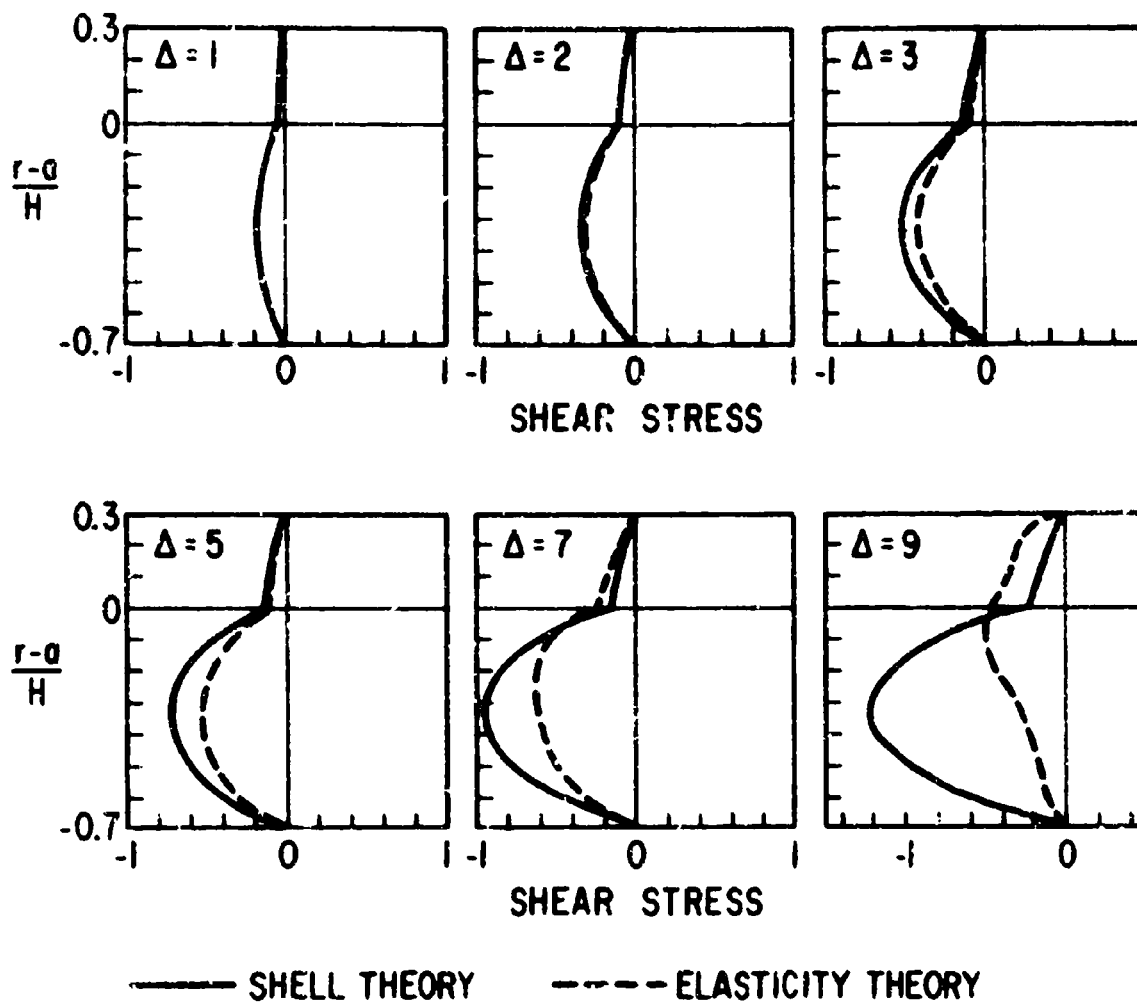


Fig. 9. First Mode Shear Stress Distributions

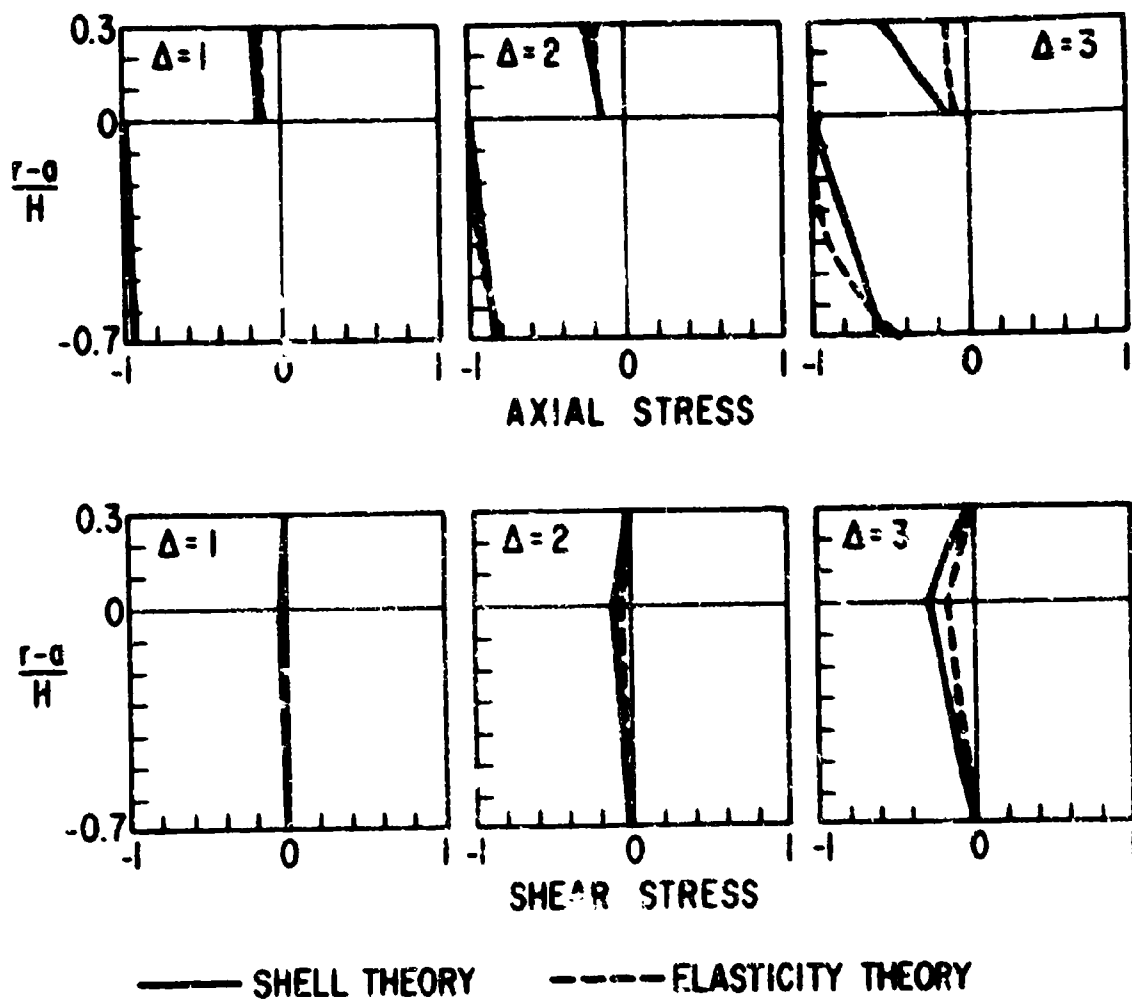


Fig. 10. Second Mode Stress Distributions

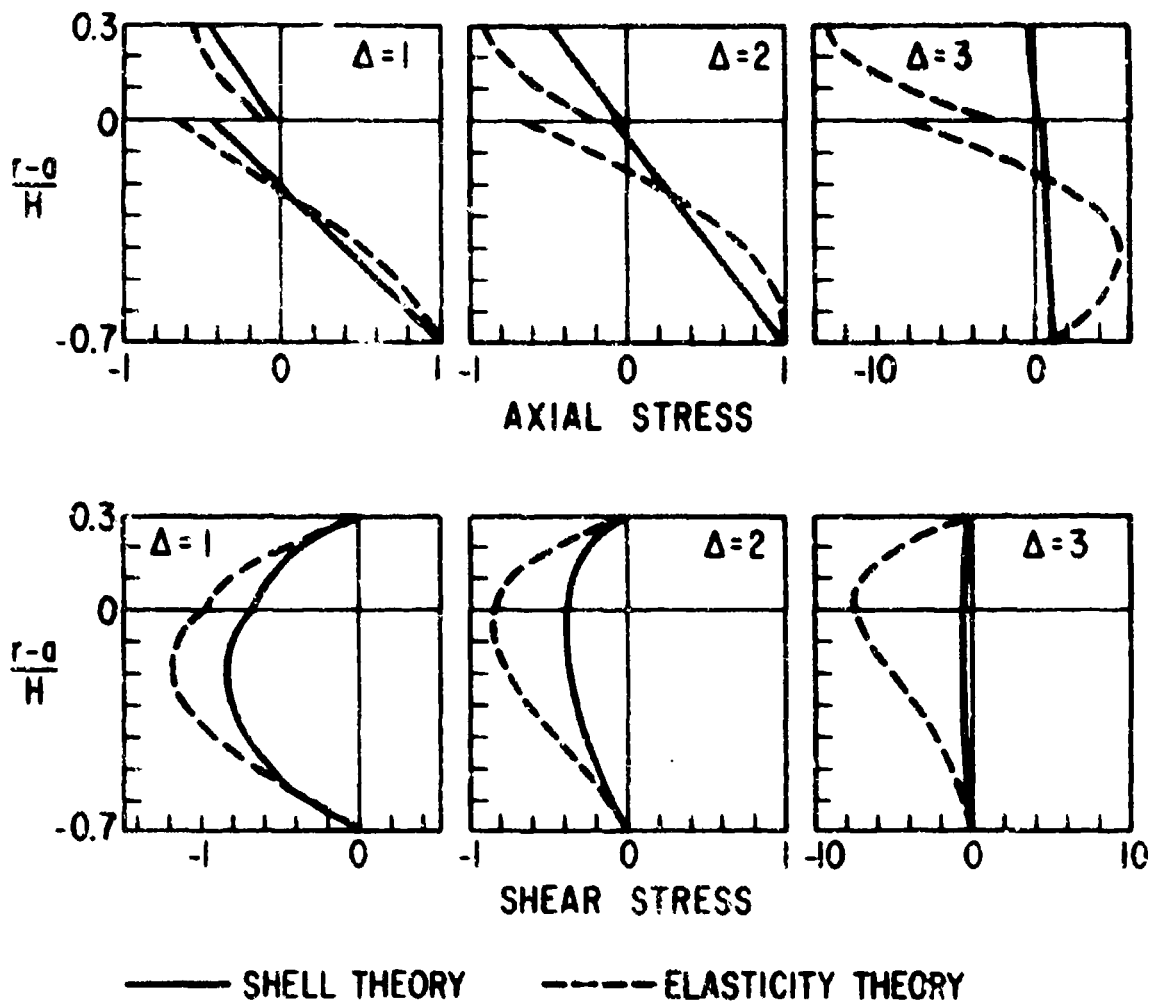


Fig. 11. Third Mode Stress Distributions

IV. CONCLUSIONS

The observations presented here on the dynamics of a two-layered cylinder are specifically pertinent to axially symmetric motions of a relatively thin-walled cylinder. The equations, however, are in no way restricted to shell-like geometry. The observations are also pertinent to the shell theory of Ref. 1 insofar as it is in agreement with the exact theory.

In many respects the results are similar to those for a homogeneous cylinder. In the lowest mode very long waves propagate with a finite phase velocity. Infinite wavelength vibrations in the second mode occur with a finite cutoff frequency corresponding to extensional ring vibrations of the cylinder. Cutoff frequencies for the higher modes correspond to simple thickness-shear or thickness-stretch vibrations. For small enough wavelengths and/or for the higher modes a good description of the motion is obtained using the simpler equations for the propagation of straight crested waves in a plate medium of the same thickness.

However, unlike a homogeneous cylinder a two-layered cylinder does not have equal surface wave velocities at its inner and outer surfaces. When the slower layer is much thinner this leads to a relative maximum at intermediate wavelength of the first mode phase velocity vs wave number curve. For larger wave numbers the phase velocity is decreasing, and it thus approaches from above the Rayleigh wave velocity of the slower medium. This character of the phase velocity dispersion relation is not matched even qualitatively by a Timoshenko-type shell theory. Presumably duplication of this behavior would require a higher order shell theory accommodating cross section distortions such that initially plane cross sections of either layer no longer remain plane in the deformed shell.

Some further conclusions pertinent to the development of higher order shell theories are in order. The displacement and stress distributions presented

here reinforce the conclusion that good results are obtained by incorporating shear deformation individually in each layer such as was done in Ref. 1. Also it appears that if one desired to extend the range of applicability of the theory of Ref. 1 by adding more dependent variables the next logical step would be incorporation of thickness-stretch deformation individually in each layer.

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REFERENCES

1. J. P. Jones and J. S. Whittier, TDR-269(4230-10)-11, Aerospace Corp., El Segundo, Calif. (30 December 1964).
2. L. Pochhammer, J.f.d. reine u. angew. Math. 81, 324 (1876).
3. C. Chree, Trans. Cambridge Phil. Soc. 14, 250 (1889).
4. J. Ghosh, Bull. Calcutta Math. Soc. 14, 31 (1923).
5. D. C. Gasis, J. Acoust. Soc. Am. 30, 786 (1958).
6. D. C. Gasis, J. Acous. Soc. Am. 31, 568 (1959).
7. D. C. Gasis, J. Acoust. Soc. Am. 31, 573 (1959).
8. J. E. Greenspon, J. Acoust. Soc. Am. 31, 1682 (1959).
9. J. E. Greenspon, J. Acoust. Soc. Am. 32, 571 (1960).
10. G. Herrmann and I. Mirsky, J. Appl. Mech. 23, 563 (1956).
11. I. Mirsky and G. Herrmann, J. Acoust. Soc. Am. 29, 1116 (1957).
12. I. Mirsky and G. Herrmann, J. Appl. Mech. 25, 97 (1958).
13. P. M. Naghdi and R. M. Cooper, J. Acoust. Soc. Am. 28, 56 (1956).
14. R. M. Cooper and P. M. Naghdi, J. Acoust. Soc. Am. 29, 1365 (1957).
15. J. F. Bird, J. Acoust. Soc. Am. 32, 1413 (1960).
16. J. F. Bird, R. W. Hart, and F. T. McClure, J. Acoust Soc. Am. 32, 1404 (1960).
17. J. H. Baltrukonis, W. G. Gottenberg, and R. N. Schreiner, J. Acoust. Soc. Am. 33, 1447 (1961).
18. H. D. McNiven, J. L. Sackman, and A. H. Shah, J. Acoust. Soc. Am. 35, 1602 (1963).
19. W. M. Ewing, W. S. Jardetsky, and F. Press, Elastic Waves in Layered Media (McGraw-Hill Book Co., Inc., New York, 1957).

20. H. Saito and K. Sato, J. Appl. Mech. 29, 287(1962).
21. J. P. Jones, J. Appl. Mech. 31, 213 (1964).
22. R. D. Mindlin. "An Introduction to the Mathematical Theory of Vibrations of Elastic Plates," prepared for U. S. Army Signal Corps Engineering Laboratories, Fort Monmouth, N. J., Signal Corps Project 142B, 1955.
23. I. Mirsky, J. Acoust. Soc. Am. 36, 41 (1964).

APPENDIX

EXPRESSIONS FOR BOUNDARY CONDITIONS

Equations resulting from the satisfaction of free surface and interface boundary conditions, Eqs. 15, appear in this Appendix; they are:

$$A_1 P_{11} + B_1 Q_{11} + C_1 R_{11} + D_1 S_{11} = 0 \quad (A-1)$$

$$A_1 T_{11} - B_1 U_{11} + C_1 V_{11} - D_1 W_{11} = 0 \quad (A-2)$$

$$A_1 \mu_1 P_{10} + B_1 \mu_1 Q_{10} + C_1 \mu_1 R_{10} + D_1 \mu_1 S_{10} = A_2 \mu_2 P_{20} + B_2 \mu_2 Q_{20} + C_2 \mu_2 R_{20} + D_2 \mu_2 S_{20} \quad (A-3)$$

$$A_1 \mu_1 T_{10} - B_1 \mu_1 U_{10} + C_1 \mu_1 V_{10} - D_1 \mu_1 W_{10} = A_2 \mu_2 T_{20} - B_2 \mu_2 U_{20} + C_2 \mu_2 V_{20} - D_2 \mu_2 W_{20} \quad (A-4)$$

$$A_1 \epsilon_1 I_1(k \epsilon_1 a) - B_1 \epsilon_1 K_1(k \epsilon_1 a) + C_1 \delta_1 I_1(k \delta_1 a) - D_1 \delta_1 K_1(k \delta_1 a) = A_2 \epsilon_2 I_2(k \epsilon_2 a) - B_2 \epsilon_2 K_1(k \epsilon_2 a) + C_2 \delta_2 I_1(k \delta_2 a) - D_2 \delta_2 K_1(k \delta_2 a) \quad (A-5)$$

$$A_1 I_0(k \epsilon_1 a) + B_1 K_0(k \epsilon_1 a) + C_1 \delta_1^2 I_0(k \delta_1 a) + D_1 \delta_1^2 K_0(k \delta_1 a) = A_2 I_0(k \epsilon_2 a) + B_2 K_0(k \epsilon_2 a) + C_2 \delta_2^2 I_0(k \delta_2 a) + D_2 \delta_2^2 K_0(k \delta_2 a) \quad (A-6)$$

$$A_{22} P_{22} + B_2 Q_{22} + C_2 R_{22} + D_2 S_{22} = 0 \quad (A-7)$$

$$A_2 T_{22} + B_2 U_{22} + C_2 V_{22} + D_2 W_{22} = 0 \quad (A-8)$$

where the lengthier combinations of Bessel functions have been abbreviated according to the following

$$\left. \begin{aligned} P_{ij} &= (1 + \delta_i^2) I_0(k\epsilon_i a) - 2\epsilon_i^2 \frac{I_1(k\epsilon_i a)}{k\epsilon_i a} \\ Q_{ij} &= (1 + \delta_i^2) K_0(k\epsilon_i a) + 2\epsilon_i^2 \frac{K_1(k\epsilon_i a)}{k\epsilon_i a} \\ R_{ij} &= 2\delta_i^2 \left[I_0(k\delta_i a) - \frac{I_1(k\delta_i a)}{k\delta_i a} \right] \\ S_{ij} &= 2\delta_i^2 \left[K_0(k\delta_i a) + \frac{K_1(k\delta_i a)}{k\delta_i a} \right] \\ T_{ij} &= 2\epsilon_i I_1(k\epsilon_i a) \\ U_{ij} &= 2\epsilon_i K_1(k\epsilon_i a) \\ V_{ij} &= \delta_i (1 + \delta_i^2) I_1(k\delta_i a) \\ W_{ij} &= \delta_i (1 + \delta_i^2) K_1(k\delta_i a) \end{aligned} \right\} \quad (A-9)$$

where i takes on values 1 and 2. The subscript j takes on values 1, 0, and 2, which corresponds to multiplying the arguments by $1 + \lambda_1$, 1, and $1 - \lambda_2$, respectively; e. g.,

$$P_{11} = (1 + \delta_1^2) I_0[k\epsilon_1 a(1 + \lambda_1)] - 2\epsilon_1^2 \frac{I_1[k\epsilon_1 \mu(1 + \lambda_1)]}{k\epsilon_1 \mu(1 + \lambda_1)}, \text{ etc.}$$

Here $\lambda_1 = (n_1/a)$ and $\lambda_2 = (n_2/a)$.

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13. ABSTRACT The linear theory of elasticity is used to investigate axially symmetric wave propagation in an infinitely long two-layered cylinder. Each material is taken to be homogeneous and isotropic. A perfect bond is assumed at the interface, while the inner and outer boundaries of the composite cylinder are treated as traction-free. The dispersion determinant relating phase velocity and wave number for a harmonic train of waves satisfying these boundary conditions is presented. The character of the dispersion equation is investigated analytically and numerically. Stress and displacement distributions are also presented for the numerical example. Comparisons are made with an approximate solution of the same problem obtained by means of a thin shell theory incorporating thickness-shear deformation of each layer.			

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