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INTERNAL BLAST DAMAGE MECHANISMS
COMPUTER PROGRAM

James F. Proctor

Naval Ordnance Laboratory

Prepared for:

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31 August 1972

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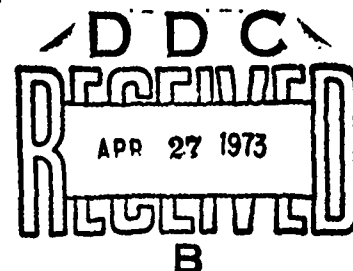
NOL

NAVAL ORDNANCE LABORATORY, WHITE OAK, SILVER SPRING, MARYLAND

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AIR/GROUND EXPLOSIONS DIVISION
EXPLOSIONS RESEARCH DEPARTMENT
NAVAL ORDNANCE LABORATORY
SILVER SPRING, MARYLAND 20910

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Internal Blast Damage Mechanisms Computer Program

The work described in this report was performed under NOL Task 594/W-PAFB, Internal Blast Mechanisms under the sponsorship of the Survivability-Vulnerability Branch, Prototype Division, Air Force Flight Dynamics Laboratory, Wright-Patterson AFB (MIPR FY1456-71-00011). The objective of this task was to develop a computer program for describing blast characteristics associated with the detonation of a high explosive projectile internal to an aircraft structure. It is expected that this program will become an item in the component damage data bank under development by the Aerial Target Vulnerability Program of the Joint Technical Coordinating Group for Munitions Effectiveness.

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This report is also available as 61 JTCG/ME-73-3.

ROBERT WILLIAMSON II
Captain, USN
Commander



C. J. ARONSON
By direction

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SUMMARY AND CONCLUSIONS

Assessment of damage to aircraft structures from the detonation of explosive projectiles internal to the aircraft requires a detailed knowledge of the dynamic pressure loads applied to various structural elements. NOL has developed a computer program that is capable of generating characteristic blast loading parameters associated with confined explosions in a form readily usable by aircraft design engineers and vulnerability analysts. Existing state-of-the-art explosion theory and experimental data were used as the basis for the shock wave calculations available in the code. An improved method of predicting the confined-explosion gas pressure that exists after shock dissipation was developed especially for this code. Any size explosion can be treated by the code for any ambient altitude condition up to and above 50,000 ft, and the code includes the blast properties of some 29 different explosives including mono, composite, and aluminized varieties.

The computer program analytically divides the internal explosion into two damaging mechanisms--the shock wave and the confined-explosion gas pressure. For the shock wave it generates the incident and normally reflected pressure-time histories and impulses for the positive phase duration at a specified distance from the explosion. Existing data and theory were used to develop the shock calculational model. The code reduces the shock calculation for all cases to the reference data from a free-field, bare, spherical 1-lb TNT explosion. Variables that affect airblast which are included in the code for establishing an equivalent TNT spherical explosion are (1) explosive weight, (2) type of explosive, (3) cylindrical charge geometry, (4) case weight of the projectile, and (5) ambient pressure and temperature at the location in the aircraft where the explosion occurs.

For an explosion internal to a confining structure or compartment, a long-duration quasi-static pressure exists after dissipation of the shock wave. The maximum value of the pressure, defined as the confined-explosion gas pressure, is dependent on these parameters; (1) weight of explosive, (2) type of explosive (chemical composition), (3) volume of compartment, and (4) pressure and temperature of the air initially in the compartment. Because of the inadequacies of existing methods of calculating the confined-explosion gas pressure, a technique was developed especially for this program that follows the energy generation of the chemical reactions and the changes in gas properties as the confined-explosion gas pressure is developed. In a completely closed compartment or structure, heat loss to the surrounding walls is the only mechanism for reducing the pressure in time, but this phenomenon is neglected in this program because of the very long durations involved. However, for the aircraft structure, there will be openings or vent areas through which the confined gases can escape such as the initial opening due to entry of the projectile into the compartment and any fragment penetration openings. Also the pressure can change abruptly due to wall failure of the compartment which introduces a new compartment volume. The computer program calculates the variation of the confined-explosion gas pressure with time for venting and such volume changes. Vent area and volume changes are controlled by input damage criteria for compartment wall failure.

In addition to the technical description of the calculational models contained in the computer program, a user's guide and complete documentation of the code are given in the text of the report and the attached appendices. Also nine sample problems are presented that demonstrate the many options and features of the program.

Although no single set of experimental data was available to compare with the overall code performance, each individual calculational model was tested against pertinent experimental data. Sufficient shock and confined-explosion gas pressure data were found for comparison, and the agreement with code predictions in these cases was excellent. It is concluded that the calculational models

for these most important aspects of the internal blast loading can be used with justifiable confidence. Whereas the pressure-time decay of the confined-explosion gas pressure due to venting has been verified with limited data, the introduction of volume changes has not been tested. The method for treating instantaneous volume changes is based on fundamental thermodynamic relations, thus there is no reason to believe that this section of the calculation detracts from the use of the code for general aircraft internal blast problems.

The computer program has wide range potential for use in studies of structural response of any military or civilian system to an internal explosion be it aircraft, naval ship, land vehicle, or building structure. Although it is adequate for the aircraft problems for which it is presently designed, there are five areas in the code that require additional study and possible modifications before its generality can be claimed for large explosions in large structure compartments such as ship compartments and building rooms. These are (1) multiple shock reflections from surrounding walls, (2) heat losses to surrounding walls that might reduce the confined-explosion gas pressure to a significant degree for large structures, (3) variable backpressure to the venting process, (4) gas pressure-time history where mixing of gases after wall failure occurs in a finite time interval, and (5) subsequent chemical reactions with the air in adjacent compartments after wall failure if complete combustion is not achieved in the initial compartment. With modification to the code reflecting the above suggested studies, it is believed this computer code could evolve as a general service-wide tool for the investigations of structural response to internal explosion loading.

CHAPTER 1
INTRODUCTION

One of the current ongoing tasks of the Aerial Target Vulnerability (ATV) Program of the Joint Technical Coordinating Group for Munitions Effectiveness (JTCC/ME) has been the development of a component damage data bank. An item defined in this data bank is the vulnerability of aircraft to internal blast from high explosive projectiles. Under the direction of the Survivability-Vulnerability Branch, Prototype Division, Air Force Flight Dynamics Laboratory (AFFDL), the objectives of this task are (1) to define the internal blast loading characteristics from a high-explosive projectile, (2) to determine the damage to aircraft structures, and (3) to assess the vulnerability of these structures to internal blast effects. The Naval Ordnance Laboratory (NOL) was assigned the technical solution of the first task problem area, namely, to define the internal blast loading characteristics.

Specifically, the objective of the NOL program was to develop mathematical and graphical techniques for describing blast characteristics associated with the detonation of a high explosive projectile internal to an aircraft structure. Existing state-of-the art experimental data and explosion theory were to be combined using sound engineering judgment to provide a computer program capable of generating characteristic shock wave and blast loading from an explosion internal to an aircraft. Execution of the computer program and the resultant loading functions were to be in a form readily usable by aircraft design engineers and vulnerability analysts.

Although this task was directed to the solution of the aircraft problem, the concepts, content, and format of the resultant computer code can be related to any military or civilian system be it aircraft, naval ship, land vehicle, or building structure. The code was structured to accommodate easy modification for any new system.

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With usage on response problems for structures other than aircraft, it is hoped that a more complete internal blast loading computer program will evolve for general use.

CHAPTER 2

GENERAL DESCRIPTION AND LIMITATIONS OF COMPUTER PROGRAM

Assume that a high explosive is detonated in a closed structure of some arbitrary geometry with a small vent opening. If a pressure sensor were to be placed on the wall of the structure, it would indicate a pressure-time history of the type shown in Figures 2.1(a) and (b). On an expanded time scale (a), one would note the initial peak reflected shock overpressure, ΔP_r , followed by subsequent reflected shock pulses from the adjacent confining wall of the structure. The oscillations would dissipate leaving a quasi-static overpressure, ΔP_g , created by the heated gases contained in the structure; this pressure is defined as the confined-explosion gas pressure. On a reduced time scale (b), the shock reflections would appear as high spikes near time zero. The confined-explosion gas pressure, ΔP_g , would be clearly established on this time scale. Even for a completely closed structure, the gas pressure would slowly decrease in time due to heat losses to the surrounding structure walls. For a structure with some openings through which venting could occur, the gas pressure would decay much more rapidly.

An accurate description of the pressure-time history during the multiple reflected shock phenomena in a closed structure of arbitrary configuration was far beyond the scope of this program effort and economically beyond the scope of any three-dimensional hydrodynamic code. For this reason the shock wave calculations in this program are limited to the incident and normally reflected pressure-time shock, depicted in Figure 2.1 (c), arriving initially at a point on the structure wall. It is believed that the normally reflected pressure-time shock history and associated reflected impulse is sufficient to provide a meaningful index in determining the local structural response to shock wave loading. Further it is believed that the predominant damaging mechanism from an internal explosion

is the confined-explosion gas pressure. For most applications, the structure wall can be treated as though it were given an initial velocity by the shock wave as an initial boundary condition. Subsequent loading on the structure is defined by the confined-explosion gas pressure which is handled as a separate loading phenomenon completely decoupled from the shock wave.

The initial magnitude of the confined-explosion gas pressure is determined from a technique developed specifically for this program, which will be described in detail in a subsequent chapter. Relative to the slower plastic response time of a typical aircraft structure, it is assumed that the confined-explosion gas pressure, ΔP_g , depicted in Figure 2.1 (d) is developed instantaneously in time. In other words, the chemical reaction of the explosion gas products with the surrounding air in the initial confines of the structure and the heat transfer to the resultant gas mixture are assumed to occur instantaneously to develop the confined-explosion gas pressure. This pressure decreases in time due to venting through available openings created by the initial entry of the projectile into the structure, subsequent openings from fragment penetrations, and any normal structural openings such as cable passageways. Venting calculations assume a constant back pressure equal to atmospheric conditions outside the aircraft since most leakage would occur through fragment penetrations in the aircraft skin. Heat losses to the structure walls that could reduce the gas pressure are neglected because (1) significant heat loss would require times much larger than the plastic response times of the structures associated with the typical small aircraft compartments and (2) pressure decreases much more rapidly from venting through even the small projectile penetration opening than from heat losses.

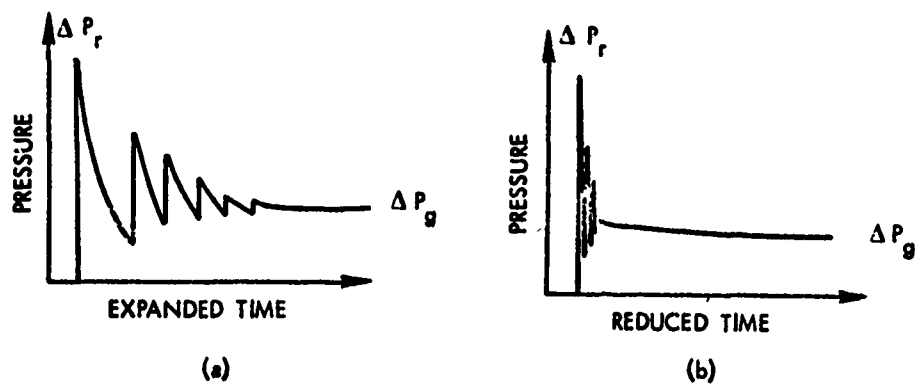
Provisions are made to account for sudden changes in gas pressure due to structural failure resulting in rapid expansion into an adjacent compartment. It is assumed for these calculations that the change in pressure is instantaneous relative to the much slower plastic response times of the aircraft structures. Also in general the small compartment sizes in an aircraft structure would indicate a rapid stabilization of pressures if compartment walls failed.

In summary to this point, four assumptions have been made that may limit the use of this computer code to aircraft only. They are: (1) multiple shock reflections are neglected--only the initial shock serves as a damage index, (2) no variation in venting back pressure occurs, (3) heat losses to structure walls are neglected, and (4) instantaneous change in pressure occurs with compartment wall failure propagation. Whereas it is believed that these assumptions do not significantly restrict the study of aircraft response to internal blast, general application of this computer program directly to other structures, such as ships and buildings, may be hampered by these assumptions. Therefore, flexibility in code construction has been provided to allow for easy and efficient modification to those sections that would be affected by alterations to these assumptions.

The basis for this entire study was existing state-of-the-art theory, analytical methods, and experimental data for explosions. When directed to the problem of internal blast, these in themselves introduce limitations in terms of applicability, and some are open to interpretation even by explosion experts. Since the prime users of this code probably will not be people with background in explosion effects, all pertinent explosion properties are self-contained in the code. Only the type and amount of explosive are required as input to the code. In calculations where limitations arising from theory and data deficiencies are encountered, caution statements are clearly indicated in the output statements for the user's benefit.

The next four chapters present the technical aspects of the calculational methods contained in the code including theoretical and experimental background, certain operational procedures, and explanations of the more important features and options in the code. Comparisons of code predictions with available experimental data are also given in these sections. Chapter 7 and the attached appendices represent a user's guide or manual for the detailed content and operation of the code. Also a number of sample problems are included in this code documentation section to acquaint the user with the various options available in the code.

TYPICAL TRACES FROM ACTUAL INTERNAL EXPLOSION



CODE APPROXIMATIONS FOR INTERNAL EXPLOSION

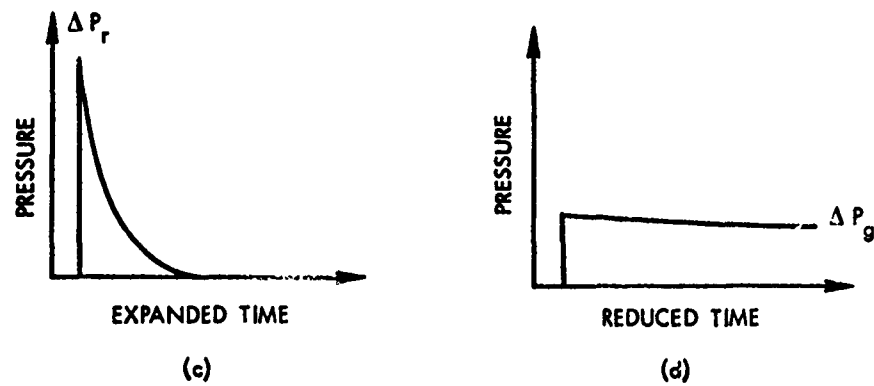


FIG. 2.1 TYPICAL PRESSURE-TIME CURVES FOR AN INTERNAL EXPLOSION

CHAPTER 3

INPUT DATA REQUIREMENTS

Explosive Parameters. For a typical high explosive projectile, only four of its properties are required as input to the computer program. They are: (1) weight of explosive, (2) type of explosive (3) length to diameter ratio of charge, and (4) metal case weight to charge weight ratio. Incorporated in the code are the pertinent properties of 24 types of explosives (Ref. (1)). Table 3.1 gives the coded properties of these 24 explosives plus 3 mono explosives that are rarely used alone as the main charge. The properties of aluminum and a common wax binder are also added. If the desired explosive is contained in this table, it is necessary only to input the index number to specify the explosive type. If one wishes to input an explosive not in the table, an index number of 0 is used and the required explosive properties must be specified. However, if for some reason the energy equivalent weight is not known, a zero for this quantity will permit shock calculations to be made with an equivalent weight of one--the same as for TNT. A diagnostic statement will appear in the output--"WFACT IS NOT KNOWN, 1.0 IS USED"--which means the shock calculations are equal to those for TNT. If the desired explosive is a mixture of components in Table 3.1, an index number of -1 is used and the weight fraction of each component must be specified. Again the energy equivalent weight may not be known, but it can be handled in the same manner as before by letting the equivalent weight equal zero. The only restriction on the type of explosive used in this program is that the explosive must be of C-H-N-O form with aluminum as the only possible metallic additive. It should be noted that this restriction arises from the confined-explosion gas pressure calculations; and as it will be discussed in subsequent chapters, this restriction has no obvious theoretical basis but must be invoked because of the lack of experimental data on other metallic additives or non-C-H-N-O explosives.

The computer program is capable of making corrections for cylindrical charge shape factors for length to diameter ratios (L/D) between 2 and 10. This is sufficiently general to accommodate most common anti-aircraft projectiles. If the L/D ratio is less than 2 as input, the code will treat the charge as spherical.

To account for effects of metal casing on the degradation of the shock wave, it is necessary to input the case weight to charge weight ratio (M/C) for the weapon. The total case weight (case, nose, fins, and fuze) should not be used in determining the ratio. Rather, it is recommended that only the case weight immediately adjacent to the explosive charge in the radial direction be used.

Initial Conditions in Structural Compartment. In order to calculate internal blast characteristics, it is necessary to specify the initial geometric properties of the structural compartment in which the explosion occurs and of the air that is confined in the compartment. Specifically for this computer program, they are (1) initial ambient pressure and temperature of the confined air (air is treated conventionally as 79% N₂ and 21% O₂ by volume), (2) initial gas volume of the compartment, (3) initial vent area for confined gases (would include opening from projectile entry and additional openings from fragment penetrations), and (4) ambient pressure or backpressure against which venting would occur, i.e., air pressure outside the compartment. If the ambient conditions of items (1) and (4) are the same as those for air at the altitude at which the aircraft is located, only the altitude of the aircraft need be specified. For this case the computer code has the 1959 ARDC standard atmosphere taken from reference (2) as a subroutine for determination of the initial ambient pressure and temperature.

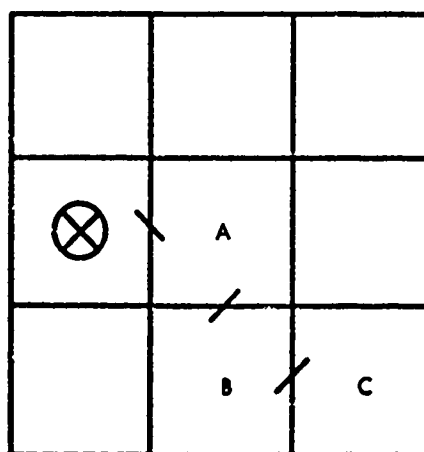
Shock Calculations. To make any shock calculation, the distance from the point of detonation to a desired location in the compartment must be designated. Specifically for this problem, this distance is measured radially from the point of detonation to the point on the structure wall where shock pressure-time information is desired. As input to the code the total number of different distances of interest must be specified, followed by a list of these desired

distances. In this manner the variation of shock loading as a function of location on a particular structural wall can be determined. There are two options in the code that must be specified as input depending on the type of shock and confined-explosion gas calculations desired. For normal code operation where both shock and confined-explosion gas pressure calculations are of interest, option 1 is used with the number and list of distances. However, if shock calculations are not desired, then option 1 is used and the number of distances is set to 0, and the shock calculation section of the program is bypassed. On the other hand, if one wishes to examine only shock calculations, option 2 is used with the number and list of distances, and the confined-explosion gas pressure section of the program is bypassed.

Volume and Vent Area Changes. It is quite possible that damage to an aircraft from an internal projectile explosion will propagate beyond the confines of the initial compartment where detonation occurs. Excessive shock loading or confined-explosion gas pressure may fail a compartment wall allowing the confined gases to propagate to an adjacent compartment. If this pressure remains excessive, additional wall failures may occur with the subsequent spread of the confined gases. As the gases propagate to different compartments and occupy larger volumes, the pressure is reduced. When the confined-explosion gas pressure has decreased to the point where wall failure does not occur, the damage propagation stops. Although a structural response code will eventually be interfaced with this blast loading code to assess the damage propagation to these pressure loads, a skeleton format is provided in this code to allow for an initial examination of the propagation phenomena. As an example, take the box structure represented in Figure 3.1 where the initial explosion occurs in the compartment with the circled X. Through wall failure (sides with cross-lines), the damage may propagate to compartments A, B, and C. It is necessary to specify certain conditions that control the damage propagation such as wall failure criteria and the amount and condition of the air in the adjacent compartments. There are three options available in the code to specify desired characteristics of compartment wall failure.

The table in Figure 3.1 is an example of failure criteria option 3. Interpretation of this input table to the computer program is as follows. (1) If the confined-explosion gas pressure in the initial compartment is above 45 psia 0.15 sec after detonation, the wall fails allowing the gases to mix with 4 ft³ of air at 14.7 psia and 20°C in compartment A and providing an arbitrary additional vent area of 0.00545 ft². (2) If the gas pressure after mixing with air in compartment A is above 20 psia 0.60 sec after detonation, the next wall fails exposing compartment B which has 4 ft³ of 14.7 psia, 20°C air and an additional arbitrary vent area of 0.00545 ft². (3) If the gas pressure after mixing in compartment B is above 19 psia, wall failure occurs involving compartment C, etc. Option 3 offers both pressure and response time control on damage criteria. If at any step the pressure is below the tabulated value at the specified time, propagation stops. For example, if the pressure is below 20 psia at 0.60 sec after detonation, damage does not propagate to compartments B or C.

Damage criteria options 1 and 2 are simplified versions of the above. Option 1 specifies only the pressure failure levels with wall failures occurring instantaneously in time. Option 2 specifies only the time of failure irrespective of pressure level. Both of these options, like option 3, require as input the volume and ambient conditions of the air in the various compartments and any additional vent area.



FAILURE PRESSURE (PSIA)	FAILURE TIME (SEC)	ADDITIONAL VOLUME (CU FT)	ADDITIONAL AREA (SQ FT)	AMBIENT PRESSURE (PSIA)	AMBIENT TEMPERATURE (C)
45	.15	4	.00545	14.7	20
20	.60	4	.00545	14.7	20
19	.80	4	0	14.7	20

FIG. 3.1 EXAMPLE OF FAILURE CRITERIA RESULTING IN VOLUME AND VENT AREA CHANGES

TABLE 3.1
LIST OF EXPLOSIVE PROPERTIES
(BASED ON DATA FROM: REF. 1)

INDEX NUMBER	EXPLOSIVE NAME	EQUIVALENT WEIGHT F_e	HEAT OF FORMATION (CAL/GM)	WEIGHT FRACTIONS OF COMPONENTS				
				C	H	N	O	AL
1	TNT	1.00	-78.40	0.370	0.022	0.185	0.423	0
2	TNETB	1.13	-307.1	0.186	0.017	0.217	0.580	0
3	EXPLOSIVE D	0.85	-386.3	0.293	0.025	0.227	0.455	0
4	PENTOLITE	1.17	-242.8	0.280	0.024	0.182	0.514	0
5	PICRATOL	0.90	-238.5	0.329	0.024	0.207	0.440	0
6	CYCLOTOL	1.14	22.79	0.225	0.026	0.320	0.429	0
7	COMP 8	1.10	4.33	0.252	0.026	0.298	0.424	0
8	RDX/WAX 98/2	1.19	57.00	0.176	0.030	0.371	0.423	0
9	COMP A-3	1.09	24.93	0.225	0.038	0.344	0.393	0
10	TNETB/AL 90/10	1.23	-276.4	0.168	0.014	0.196	0.522	0.100
11	TNETB/AL 78/22	1.18	-239.5	0.146	0.012	0.170	0.452	0.220
12	TNETB/AL 72/28	1.18	-221.1	0.134	0.011	0.157	0.416	0.280
13	TNETB/AL 65/35	1.23	-199.6	0.121	0.010	0.142	0.37	0.350
14	TRITONAL	1.07	-62.72	0.296	0.018	0.148	0.358	0.200
15	RDX/AL/WAX 88/10/2	1.30	50.38	0.160	0.027	0.333	0.580	0.100
16	RDX/AL/WAX 78/20/2	1.32	43.76	0.144	0.024	0.295	0.337	0.200
17	RDX/AL/WAX 74/21/5	1.30	29.36	0.163	0.027	0.280	0.320	0.210
18	RDX/AL/WAX 74/22/4	1.30	33.28	0.154	0.026	0.280	0.320	0.220
19	RDX/AL/WAX 62/33/5	1.19	21.42	0.143	0.024	0.235	0.268	0.330
20	TORPEX II	1.24	-3.57	0.216	0.021	0.233	0.350	0.180
21	H-6	1.27	-12.56	0.223	0.025	0.224	0.318	0.210
22	H8X-1	1.21	-22.93	0.249	0.026	0.221	0.334	0.170
23	H8X-2	1.16	-21.83	0.200	0.022	0.171	0.257	0.350
24	TNETB/RDX/AL 39/26/35	1.24	-102.6	0.115	0.013	0.184	0.338	0.350
25	ALUMINUM	0	0	0	0	0	0	1.000
26	WAX	0	-392.0	0.856	0.144	0	0	0
27	RDX	0	66.16	0.162	0.027	0.379	0.432	0
28	PETN	0	-407.1	0.190	0.026	0.177	0.607	0
29	TETRYL	0	16.26	0.293	0.017	0.244	0.446	0

CHAPTER 4

SHOCK WAVE CALCULATIONS

Base Data. The principal thesis of the shock wave calculations in this computer program is that a cased, cylindrical charge of a given type and amount of explosive detonated at any altitude from sea level to at least 50,000 ft can be equated to a free-field 1-lb TNT spherical explosion at sea level. Generally explosion data given in handbooks for TNT do not provide sufficient information to yield the pressure-time history of the shock at a specified distance from the explosion; usually only peak pressures, positive phase durations, and positive impulses are given. One must turn to various hydrodynamic codes to obtain such information in lieu of extensive experimental data. However, such codes are lengthy and expensive to run and do not lend themselves to the objective of this program. It was decided to use results from a current version of the WUNDY hydrocode developed at NOL and described in reference (3), to normalize these results to form a family of pressure-time curves for a large number of distances, and to find the best empirical fit to represent these results.

Figure 4.1 shows four representative curves developed by WUNDY that demonstrate the incident pressure-time behavior of a free-field shock wave as a function of distance, R . From some 25 curves of this type that exist outside the explosion gas contact surface, it was found that the family could be represented quite well by the equation.

$$\pi = \Delta P / \Delta P_1 = (1 - \tau) e^{-\tau} \left(1 + \frac{\sigma}{A + \tau} \right) \quad (4.1)$$

where

$$\tau = (t - t_a) / t_d$$

$$\sigma = (228/R) - 0.95$$

$$A = 0.5$$

and ΔP_1 = peak incident shock overpressure

ΔP = instantaneous overpressure

t = time measured from detonation

t_a = arrival time of shock measured from detonation

t_d = positive phase duration of incident shock pulse

R = distance from detonation (cm)

From the above equations, it is seen that peak incident pressure, arrival time, and positive phase duration for a given distance are the only parameters required for development of the shock pressure-time curve. Values of peak incident overpressure and arrival time are readily available from WUNDY code results and have been tabulated in the computer program for 108 distances ranging from the charge surface to 2.342×10^6 cm (based on 1-lb bare TNT sphere). Positive phase durations for distances outside the contact surface are also obtained from WUNDY. However, inside about 70 cm, the positive phase duration of the shock wave is not completed before interaction with the contact surface occurs. Although WUNDY follows the contact surface boundary, the available runs do not yield usable results inside the contact surface. Therefore, experimental data from references (4) and (5) were used to derive approximate positive phase durations inside the contact surface. The computer code assumes that equation (4.1) continues to hold inside the contact surface. Arrival time of the contact surface is programmed into the code. When shock information is desired at a point inside the contact surface, the user is alerted to the fact that the pressure-time history is an approximation by a diagnostic or warning statement that appears in the code output--"CAUTION--CONTACT SURFACE HAS ARRIVED. DATA ARE CRUDE BEYOND T(MSEC) AFTER SHOCK ARRIVAL =".

As indicated previously, parameter values are tabulated in the computer code for 108 distances. An interpolation method was needed to accommodate any given distance. Plots of peak pressure, shock arrival time, positive phase duration, and contact surface arrival time as functions of distance on log-log scales demonstrated a nearly linear slope over relatively small intervals. Therefore, linear interpolation between the log values of the parameters and the log values of distance is coded into the program as an accurate interpolation method.

Scaling Equations. To relate a spherical TNT explosion at altitude to a 1-lb spherical TNT explosion at sea level, conventional Sachs scaling was used in the computer program. (Sachs scaling method can be found in many references on airblast from explosions, such as reference (6).) The scaling relations, as they are used in this computer code, are given as

$$R_s = R_a (W_s/W_a)^{1/3} (P_a/P_s)^{1/3} \quad (4.2)$$

$$\Delta P_a = \Delta P_s (P_a/P_s) \quad (4.3)$$

$$t_a = t_s (W_a/W_s)^{1/3} (P_s/P_a)^{1/3} (T_s/T_a)^{1/2} \quad (4.4)$$

$$I_a = I_s (W_a/W_s)^{1/3} (P_a/P_s)^{2/3} (T_s/T_a)^{1/2} \quad (4.5)$$

where

R = distance

W = charge weight

ΔP = overpressure

P = ambient pressure

t = time

T = ambient temperature

I = impulse

s = subscript denoting 1-lb TNT sphere at sea level

a = subscript denoting TNT sphere at altitude

Following the order of these equations, for a given distance, R_a , from a spherical TNT charge, W_a , at altitude pressure, P_a , and temperature, T_a ; a scaled distance, R_s , from a 1-lb TNT spherical charge ($W_s=1$) at sea level pressure and temperature, P_s and T_s , is determined. The code then calculates a pressure-time curve for the scaled distance and 1-lb TNT charge at sea level from equation (4.1) and numerically integrates this curve to determine the incident impulse. With values P_s , t_s , and I_s , equations (4.3)--(4.5) define the values of these parameters at the desired altitude. More details of this method will be presented in subsequent sections.

Equivalent Weight. In the previous section on scaling, all charges were TNT spheres. Since typical projectile charges are cased, cylindrical, non-TNT explosives, methods for equating the blast effects of a real projectile to those of an idealized TNT sphere were required. Virtually all studies directed at establishing equivalent weights have been conducted at sea level conditions. The assumption was made for this program that the relative performance of explosives is essentially the same at sea level as at altitude so that equivalent weights do not vary with altitude. The equivalent weight relating various explosive compositions in bare spherical charge form to airblast performance is defined in this report as the energy equivalent weight, f_e , and is given as an explosive property in Table 3.1. Factors relating cylindrical to spherical charges and cased cylindrical to bare cylindrical charges have been shown experimentally to depend on the peak incident overpressure level in a gross sense. Therefore, before evaluating these factors, an estimate of the peak incident overpressure for a spherical explosion scaled to sea level is made based on the energy equivalent weight alone.

First, methods were developed to determine the cylindrical charge equivalent weight. A compilation of data for bare Comp B cylindrical charges was taken from reference (7) for L/D ratios between 2 and 10. It was found that the data formed the three curves shown in Figure 4.2. The 90° curve gives peak incident overpressures measured along a line perpendicular to the longitudinal axis of the cylindrical charges; the 45° curve gives overpressures along a line inclined 45° from the longitudinal axis; and the 0° curve gives overpressures along the extension of the longitudinal axis. The orientation of the projectile with respect to the aircraft compartment structure would vary considerably depending on the mode of attack. Since the design of aircraft to withstand internal blast is of utmost concern for this program, a conservative assumption is to use the curves yielding the highest pressure. In Figure 4.2 this is the 90° curve up to a scaled distance of about 8 and then the 45° curve for scaled distances greater than 8, or the resultant composite cylindrical charge curve shown in Figure 4.3. (If one is interested in weapon selection for damaging aircraft, he would chose the composite curve of 45° and 0° in Figure 4.2.)

From the same DRI study in reference (7), bare spherical charges of the same explosive Comp B were detonated yielding the spherical charge curve in Figure 4.3. Choosing a particular pressure level, one can determine the cylindrical charge equivalent weight (f_s = weight of sphere/weight of cylinder) for equal distances. In this manner the low-pressure curve shown in Figure 4.4(a) was developed. Empirical relations that represent this curve

$$0 \leq \Delta P_1 \leq 20 \quad ; \quad f_s = 1.45 \quad (4.6)$$

$$20 \leq \Delta P_1 \quad ; \quad f_s = 0.613 (\Delta P_1)^{0.287} \quad (4.7)$$

have been programmed into the computer code. It is assumed in the program that all explosives follow the behavior of this experimental data for Comp B. Lack of complete sets of data for other explosive compounds makes this assumption necessary.

In Figures 4.2, 4.3, and 4.4, the experimental data do not extend to overpressures above 100 psi. To limit shock calculations to this experimentally verified range is too restrictive for a realistic problem where the projectile will generally be relatively close to a structure wall where incident overpressures above 100 psi will surely exist. It is improper to assume that the cylindrical charge equivalent weight will continue to increase indefinitely with pressure as given by equation (4.7). As the distance from the cylindrical charge decreases, at some point the charge will begin to appear as an infinitely-long charge or a line charge, and the equivalent weight will begin to decrease. Since no experimental data are available to provide guidance for determining equivalent weights for pressures above 100 psi, the following method is assumed. From theoretical work on line charges developed by Kirkwood and Brinkley in reference (8), the high-pressure curve shown in Figure 4.4(b) was determined. Note that the ordinate is not equivalent weight as in Figure 4.4(a), but rather a term defined for this report as comparative weight index. From the L/D ratio of the charge, this index can be converted to the equivalent weight, f_s . If one makes this conversion over the entire curve, a family of curves is generated and shown as the various L/D curves in Figure 4.5. The transition from a line charge to a cylindrical

charge of finite length is made by extending the low-pressure curve in Figure 4.4(a), or equation (4.7), until it intersects this family. Thus Figure 4.5 represents the composite example of the method used by the computer to calculate the cylindrical charge equivalent weight. Although the trend of the assumed method agrees with the expected physical behavior of cylindrical charges, any shock calculations based on this method must be viewed as approximations. Because of the uncertainty associated with this approach and the lack of experimental verification of pressure data above 100 psi for cylindrical charges in general, a diagnostic or warning statement in the computer output appears--"CHARGE SHAPE CORRECTION IS CRUDE. PSI EXCEEDS RANGE OF EXPERIMENTAL DATA".

Secondly, a method to determine the effects on airblast shock of a metal casing surrounding a cylindrical charge was required, i.e., a casing equivalent weight, f_c , to relate cased and bare cylindrical charges. As found in reference (9), a number of methods have been proposed and are given in Figure 4.6. These curves alone give no insight to the best approximation of the casing effects. Assorted case effects experimental data taken from references (10)–(13) have been plotted in Figure 4.6. Whereas it appears that the equation

$$f_c = 0.20 + 0.80/(1 + M/C); M/C = \text{case weight/charge weight}$$

best fits the experimental data, it is noted that it slightly underestimates the effects of the case for much of the data. Consistent with previous assumptions for the cylindrical charge equivalent weight, the most conservative method was sought, i.e., the method that yields the greatest pressures. This was accomplished by combining the upper two curves into one as plotted in Figure 4.7 with a replot of the experimental data points. It is noted that only one data point lies above this curve. Therefore, the casing equivalent weight used in the computer program is expressed as

$$0 \leq M/C \leq 0.53$$

$$f_c = \frac{1 + (M/C)(1-M')}{1 + M/C} = 1 - (M/C)^2/(1 + M/C) \quad (4.8)$$

$$(M' = M/C \text{ for values of } M/C \text{ less than } 1)$$

$$0.53 \leq M/C$$

$$f_c = 0.47 + 0.53/(1 + M/C) \quad (4.9)$$

(Again, if one were interested primarily in weapon selection rather than aircraft design, the lower curves of Figure 4.6 would be more suitable.) It should be pointed out that the experimental data used for casing effects were based on measured incident overpressures less than 100 psi. Therefore, a warning statement appears in the computer output to alert the user to the approximate nature of the data for overpressures above 100 psi--"CASE WEIGHT CORRECTION IS CRUDE. PSI EXCEEDS RANGE OF EXPERIMENTAL DATA".

From the above discussions, a cylindrical, cased, non-TNT explosive charge can be related to a bare spherical TNT charge by the expression

$$W_{TNT} = W \times f_e \times f_s \times f_c \quad (4.10)$$

where

W_{TNT} = weight of equivalent bare spherical TNT charge

W = weight of cased charge

f_e = energy equivalent weight from Table 3.1

f_s = cylindrical charge equivalent weight from Figure 4.4 or 4.5

f_c = casing equivalent weight from equations (4.8) and (4.9)

Free-Field Incident Pressure-Time and Impulse. Using equation (4.10) and the scaling equation (4.2), the computer can scale a high explosive projectile explosion at altitude to a free-field, bare, spherical 1-lb TNT explosion at sea level. For a specific scaled distance, the computer selects appropriate free-field explosion data required for the pressure-time equation (4.1). It then calculates incident free-field overpressures that correspond to equal time steps during the positive phase duration of the shock wave. The number of equal time steps, k , can be varied from 10 to 40; however, the number should be as large as conveniently possible because these time steps control the numerical integration procedure that calculates the positive impulse as seen by the following equation

$$I = \sum_{i=0}^{i=k} [\Delta P(i) + \Delta P(i-1)] \Delta t / 2$$

$$\Delta t = t_d / k$$

where

- I = incident impulse
- ΔP = incident overpressure
- i = step index
- Δt = time interval
- t_d = positive duration
- k = number of time steps

For the convenience of the user, time is measured both from the instant of detonation and from the instant of shock arrival at the desired distance from the explosion. The tabulated incident pressure-time information and the incident impulse are then scaled to the actual conditions at altitude using the scaling equations (4.3)--(4.5).

Normally Reflected Pressure-Time and Impulse. As stated in the general program description, accurate analysis of shock reflection in a structure of an arbitrary configuration is presently beyond the scope of this code. Normally reflected pressure-time information and normally reflected impulse have been chosen as loading indices for studying structural response to shock loading. For the small structural compartments in aircraft, shock loading from the relatively small explosive charges in anti-aircraft projectiles would be completed before any appreciable plastic response of the structure has occurred. Therefore, the shock basically can be treated as an impulsive load on an aircraft compartment structure, and it is believed that the normally reflected explosion data developed by the computer code will be sufficient to index the response of the structure to shock loads. While this assumption is sufficient, in all probability, when applied to small aircraft compartment structures, it might prove restrictive and limiting for code application to response problems

relating to large structures such as ship compartments or building rooms.

Methods for predicting the peak normally reflected overpressure have been developed and verified by experimental data over a wide range of pressure. The method used in this computer program is based on the reflection factor curve developed by Brode in reference (14) and shown in Figure 4.8 as the solid curve. With sufficient accuracy this curve is approximated by the combination of dashed curves shown in this figure. In the computer program the following equations are used to calculate reflection factors for the peak normally reflected overpressure at sea level conditions.

$$0 \leq \Delta P_1 \leq 200 \text{ psi}$$

$$f_R = 2 \left[\frac{(7)(14.7) + 4 \Delta P_1}{(7)(14.7) + \Delta P_1} \right] \quad (4.11)$$

$$200 < \Delta P_1 \leq 10,000 \text{ psi}$$

$$f_R = -3.18 + 3.97 \log_{10} (\Delta P_1) \quad (4.12)$$

$$10,000 < \Delta P_1$$

$$f_R = 13 \quad (4.13)$$

where ΔP_1 = peak incident shock overpressure.

The major problem now is how to relate this normal reflection factor derived for the peak reflected overpressure to the entire reflected pressure-time history and reflected impulse. Whereas hydrocodes exist that will follow the normally reflected shock phenomena in time, they do not lend themselves economically for use with this program. For the lack of a better method at this time, it is assumed that an adequate approximation to the reflected pressure-time history is found by multiplying the pressure level of the previously calculated incident pressure-time history by the reflection factor, f_R , derived for the peak reflected pressure. Thus the computer program multiplies the incident pressures as they are determined for the 1-lb TNT sphere at sea level by the appropriate

reflection factor, f_R , from equations (4.11)--(4.13) using the peak incident peak overpressure. It then scales these results to altitude conditions in the same manner as it scales the incident pressures. Likewise, the incident impulse is multiplied by the same reflection factor to obtain the normally reflected impulse.

If one wishes to evaluate a reflection condition other than normal, it is possible to modify the incident pressure-time curve with a reflection factor for an angle of incidence other than normal (90°). Such reflection factors can be found in Figure 4-6 of reference (15) and in Figure 3.71b of reference (16). With the uncertainties involved in this approach to reflected pressure-time histories, caution should be exercised in using the referenced factors for other angles of incidence with this method except to serve as an index or for scoping calculations.

Comparisons with Experimental Data. Since normally reflected pressure-time and impulse information are assumed to be the important shock characteristics in terms of aircraft compartment structure damage, it is most important that the code predicts normally reflected shock phenomena accurately. Unfortunately, documentation of experimental programs studying reflected shock data relatively close to the explosion has been difficult to find. The best available set of data was found in the BRL study reported in reference (5). This report presented experimental curves for peak reflected pressure and reflected impulse based on old and new tests with bare, spherical pentolite charges.

These curves are shown in Figures 4.9 and 4.10 along with calculated results from the computer program depicted by the circles. Peak reflected pressure predictions agree remarkably well with the experimental curve in Figure 4.9. Reflected impulses agree well with the experimental curve in Figure 4.10, but the relative variation is not as good as for peak reflected pressure. In all fairness to the computer program, a close study of the spread of experimental data from which the impulse curve was drawn (Figure 8 of reference (5)) reveals experimental variation as large as that observed from the code prediction-experimental curve comparison. It is important to

note here that these experimental data extend to the very high pressure range and that the reflected impulse for some of the tests include effects inside the contact surface--all questions of uncertainty in the formulation of the code. Therefore, it must be concluded that the computer code yields predictions that agree remarkably well with this set of experimental data, which certainly provides confidence to the use of the computer program for shock calculations.

Additional confidence is gained from a comparison of a reflected pressure-time trace from one of the BRL experiments with a code predicted reflected pressure-time history. Figure 6 of reference (5) gives an enlargement of a reflected pressure-time trace from a 1/8-lb pentolite test at a scaled distance of $2.5 \text{ ft/lb}^{1/3}$. This curve, which is free of extraneous noise and oscillations, is shown in Figure 4.11 as the solid curve. Shown as the dashed curve is the predicted reflected pressure-time results from the computer code. Agreement has to be classified as excellent in light of all the simplifying assumptions used in the code for shock calculations.

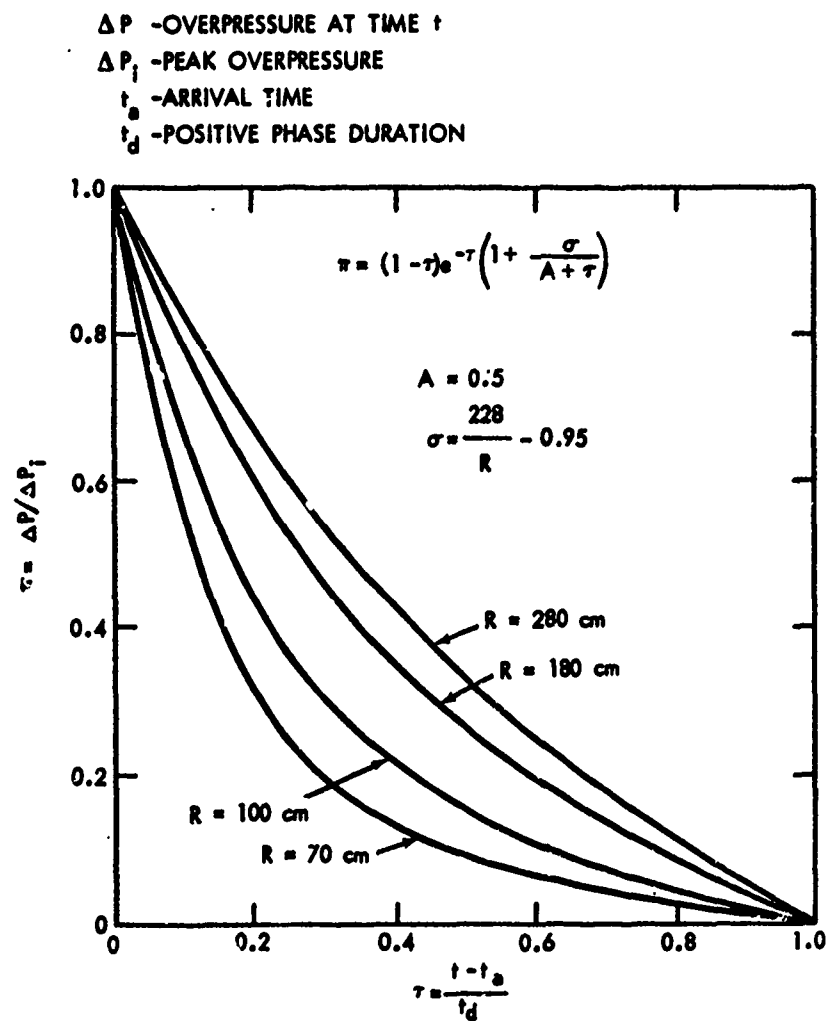


FIG. 4.1 SHOCK PRESSURE-TIME WAVE FORMS

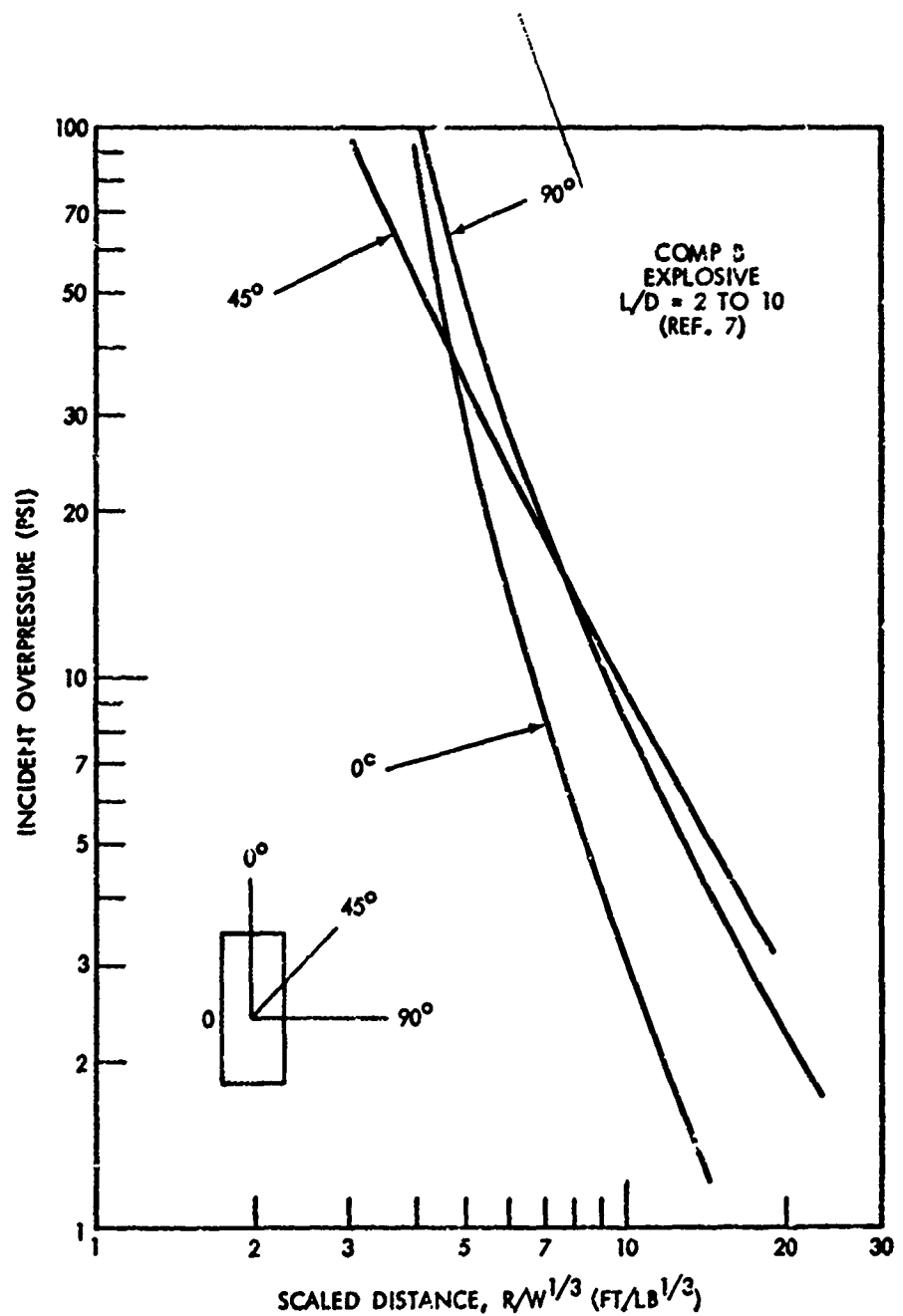


FIG. 4.2 PRESSURE-DISTANCE CURVES FOR VARIOUS ANGLES FROM AXIS OF CYLINDRICAL CHARGE

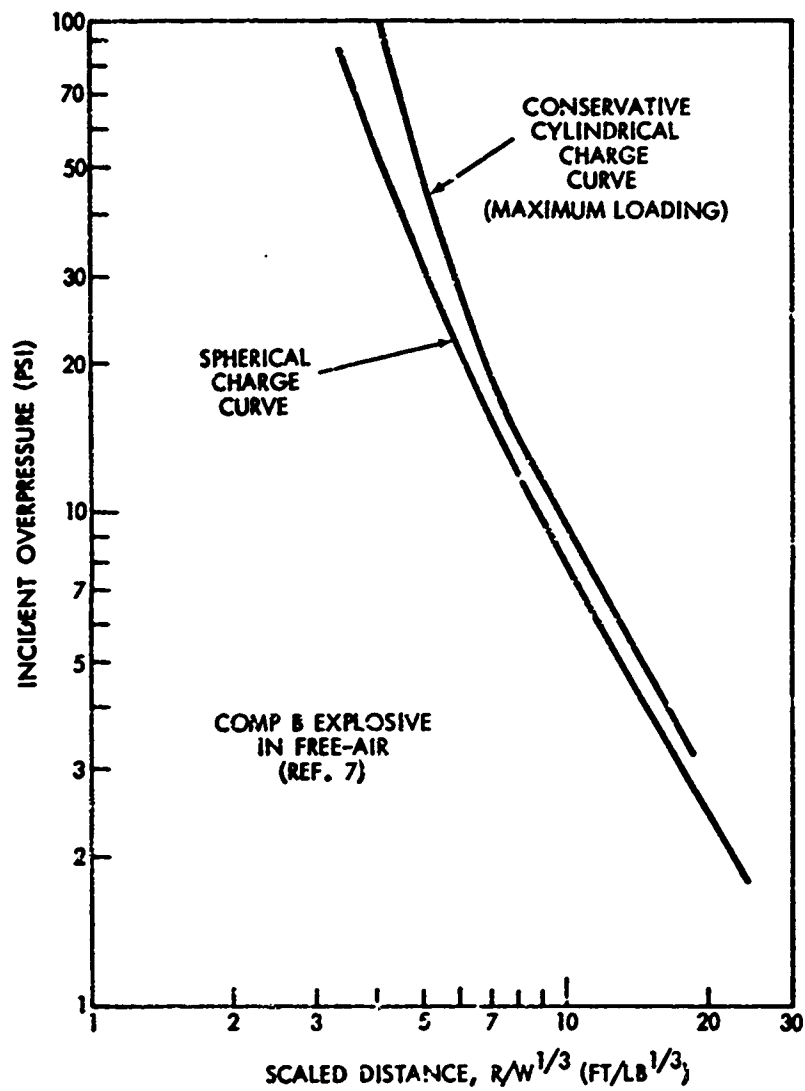


FIG. 4.3 COMPARISON OF PRESSURE-DISTANCE CURVES FOR CONSERVATIVE CYLINDRICAL AND SPHERICAL CHARGES

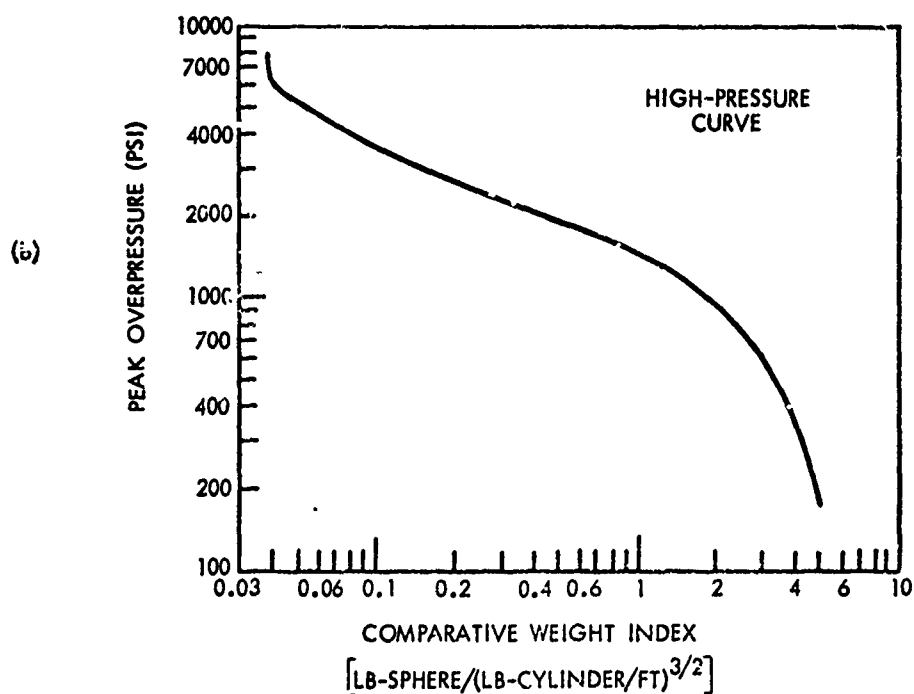
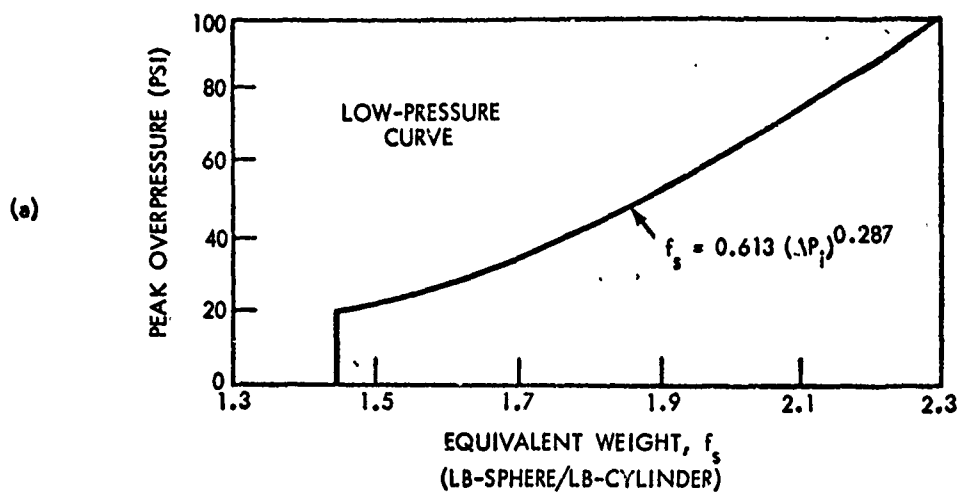


FIG. 4.4 CYLINDRICAL CHARGE EQUIVALENT WEIGHT

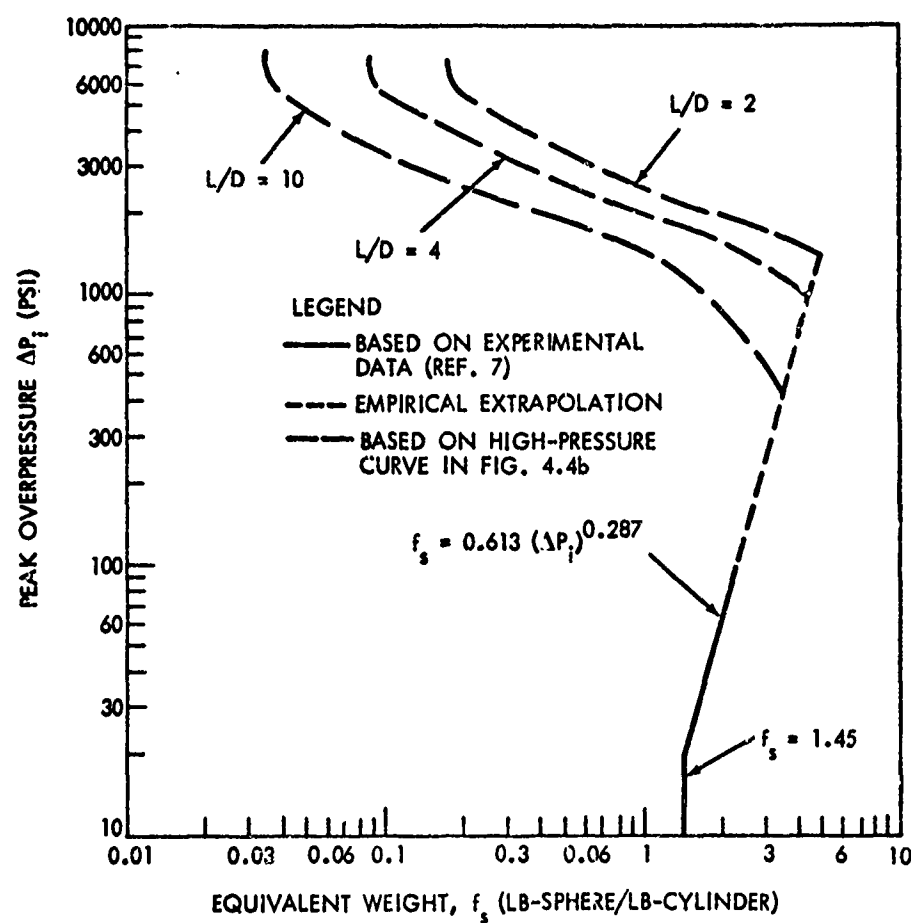


FIG. 4.5 COMPOSITE EXAMPLE OF CYLINDRICAL CHARGE EQUIVALENT WEIGHT

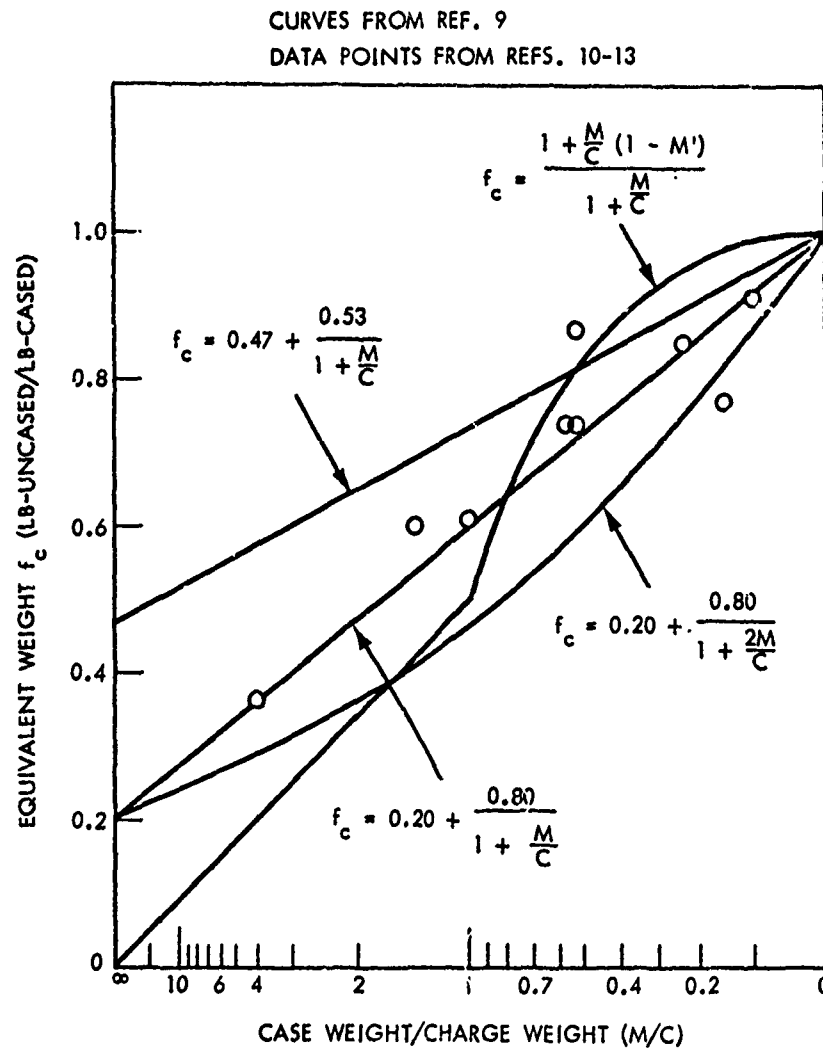


FIG. 4.6 VARIOUS METHODS FOR PREDICTING CASE EFFECTS

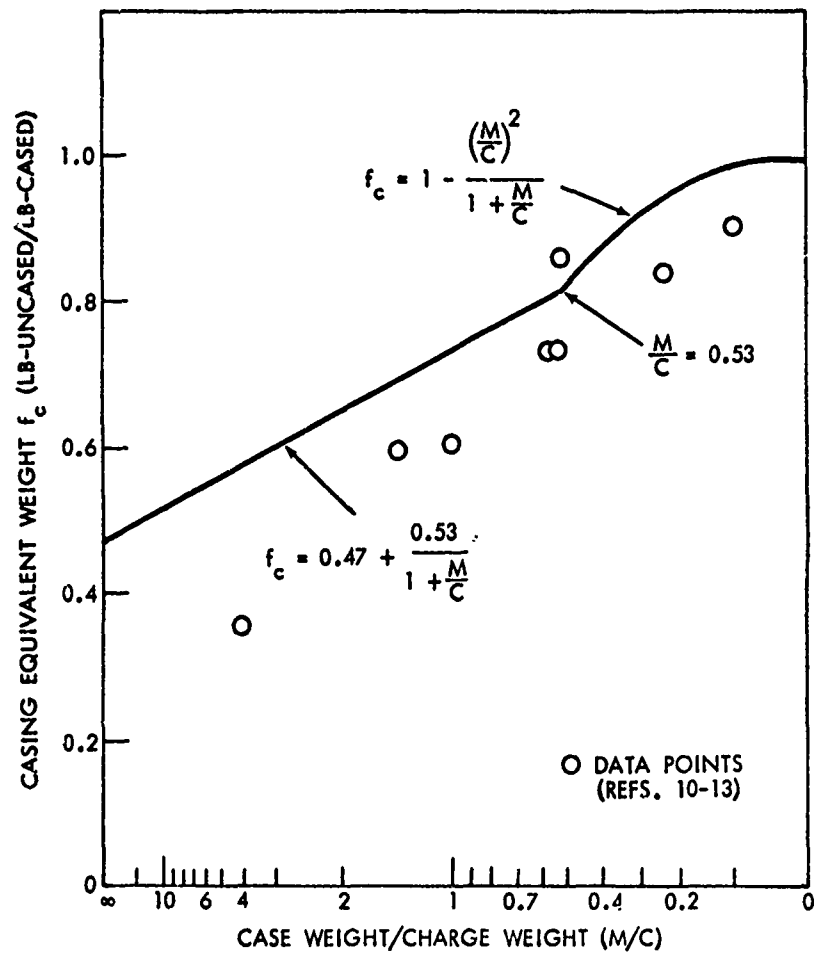


FIG. 4.7 CASING EQUIVALENT WEIGHT (MAXIMUM LOADING)

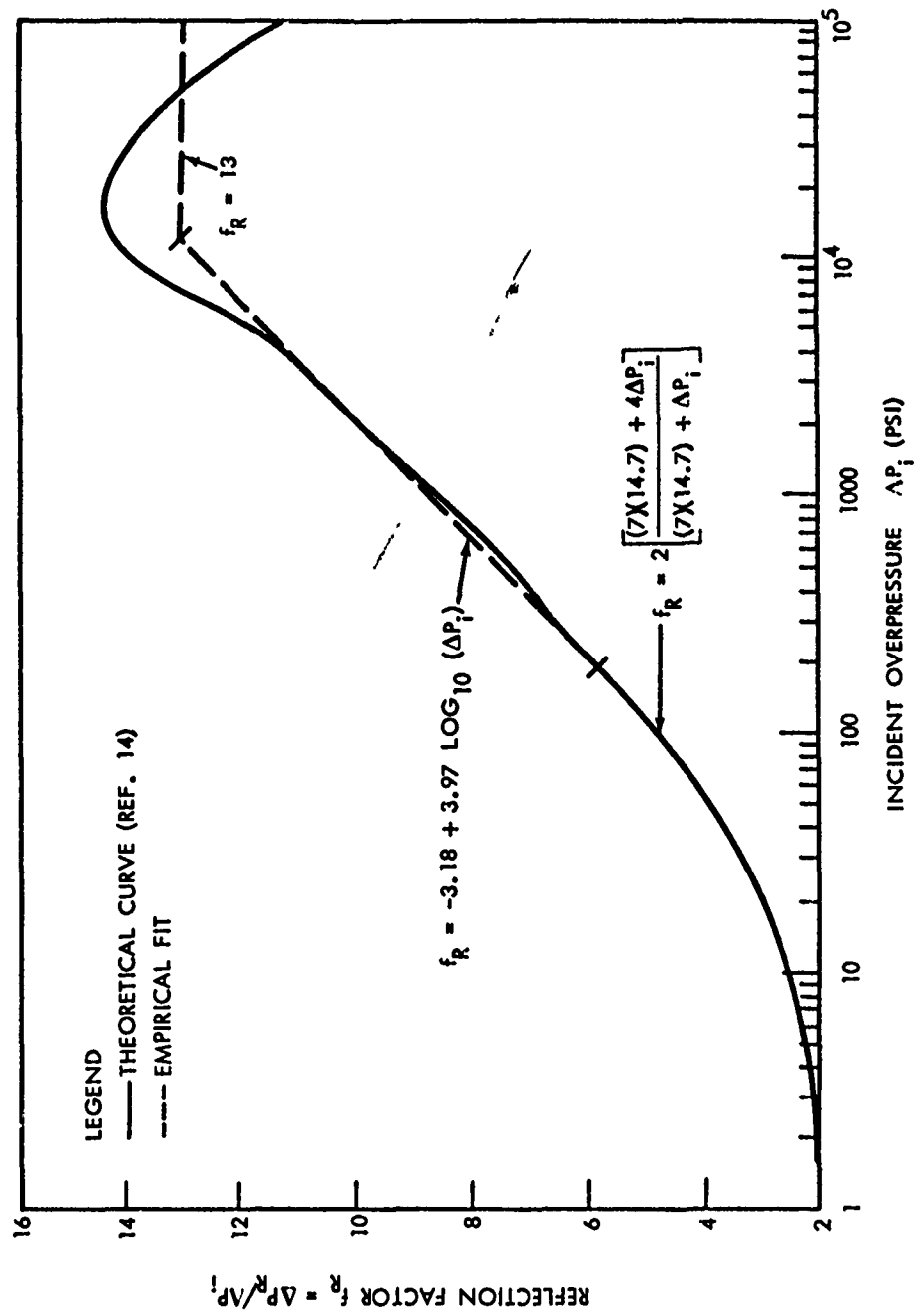


FIG. 4.8 NORMAL REFLECTION FACTORS

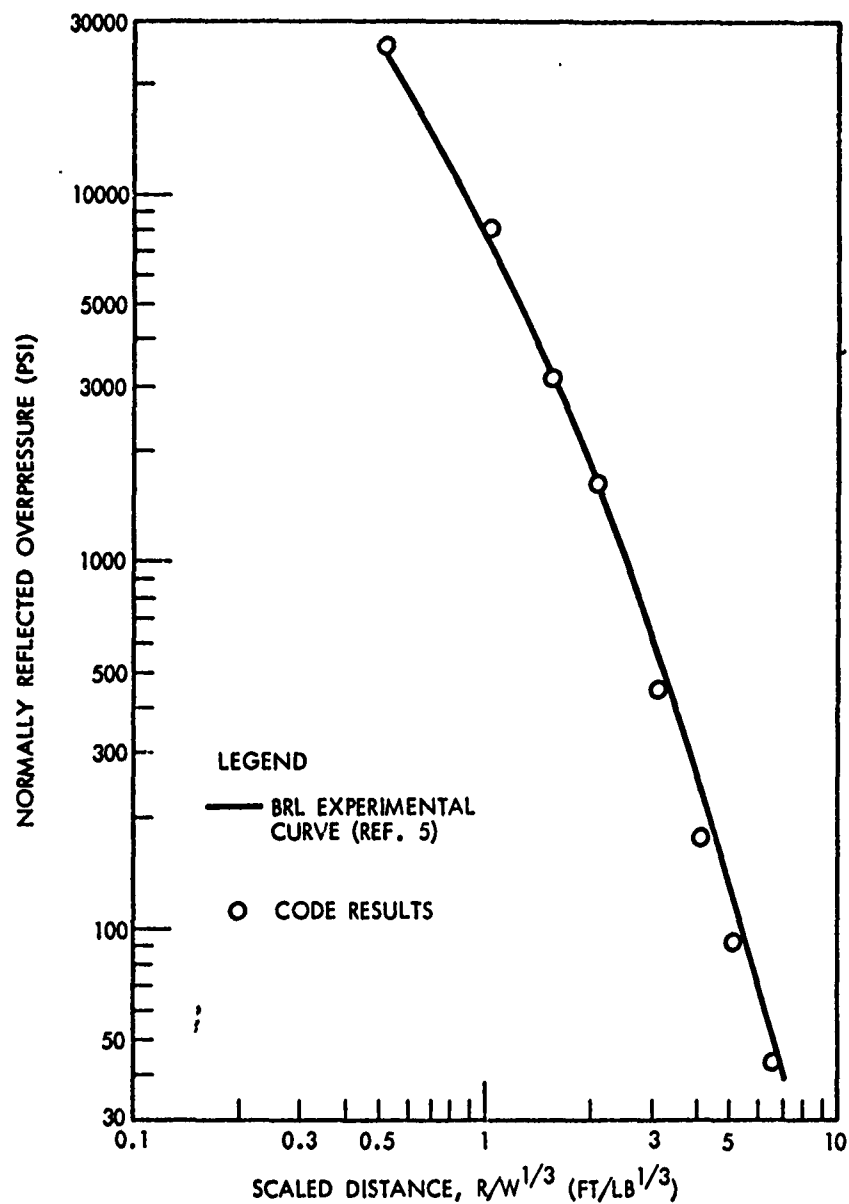


FIG. 4.9 REFLECTED PRESSURE-DISTANCE CURVE FOR PENTOLITE IN FREE-AIR AT SEA LEVEL

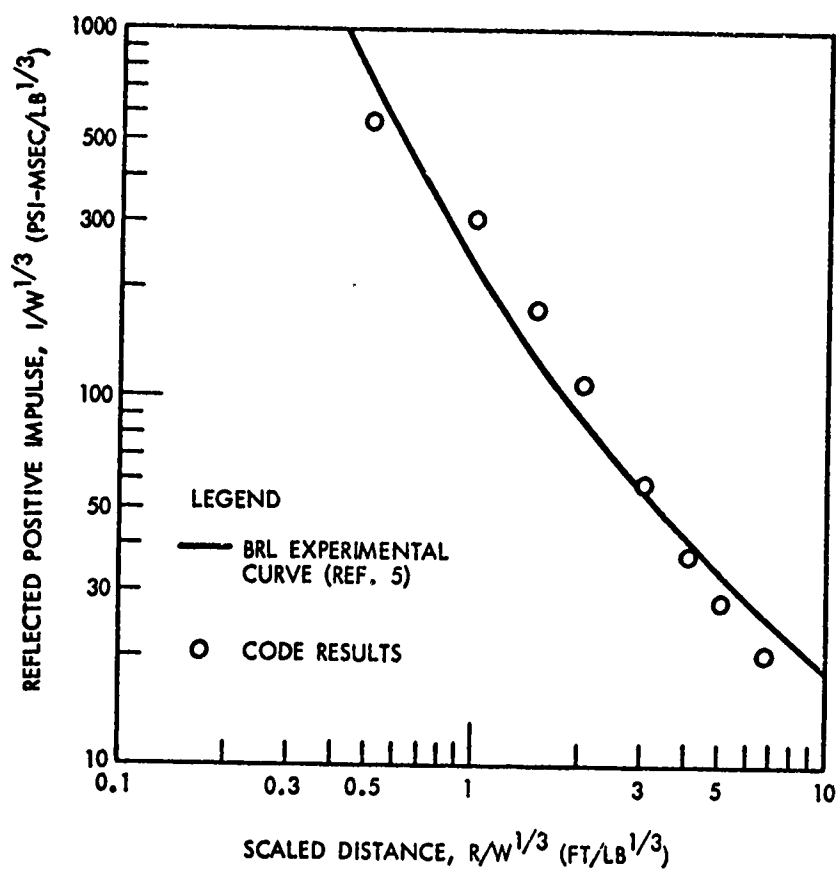


FIG. 4.10 REFLECTED POSITIVE IMPULSE-DISTANCE CURVE FOR PENTOLITE IN FREE-AIR AT SEA LEVEL

1/8-LB PENTOLITE, DISTANCE OF 2.5 FT/LB^{1/3}, FREE-AIR AT SEA LEVEL

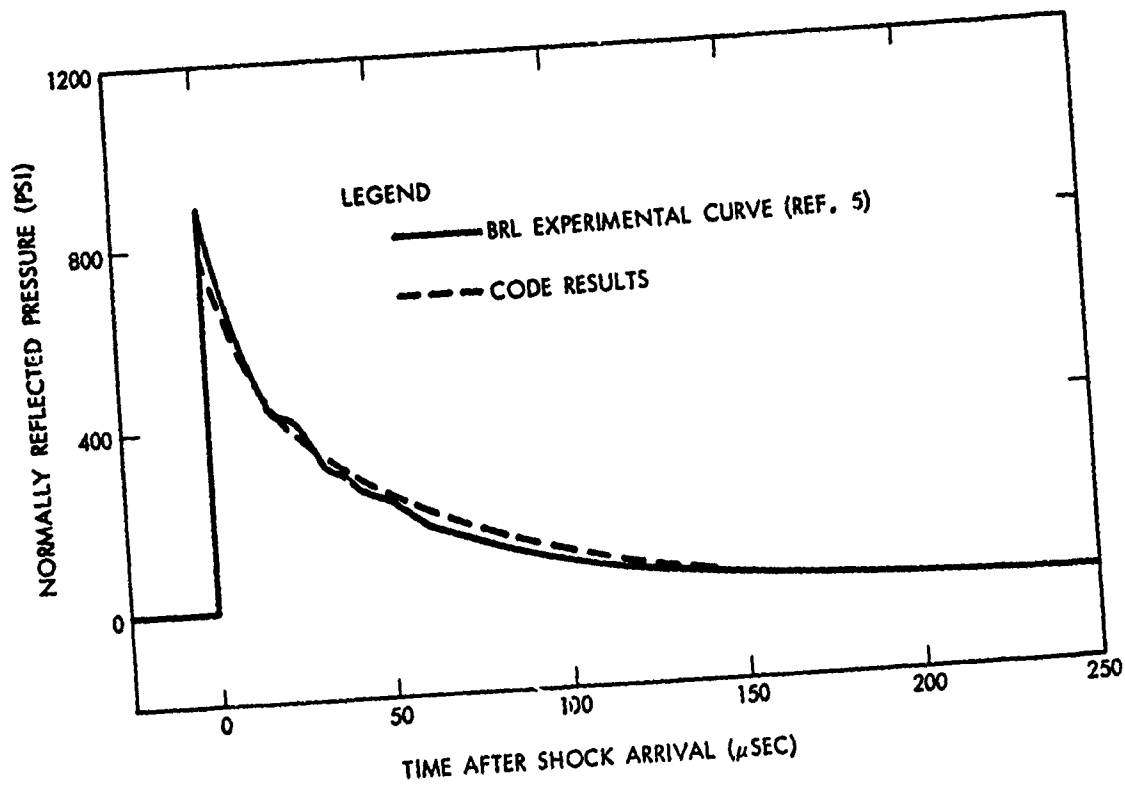


FIG. 4.11 COMPARISON OF CODE AND EXPERIMENTAL SHOCK PRESSURE-TIME HISTORIES

CHAPTER 5

CONFINED-EXPLOSION GAS PRESSURE CALCULATIONS

Phenomena Description. The development of the quasi-static pressure that exists in a closed structure after an explosion is presented in detail in references (17) and (18). It is briefly discussed here. After the multitude of shock reflections from an explosion in a completely closed structure have dissipated, there exists a significant overpressure in the structure. A tremendous amount of heat is released from the chemical decomposition of the explosive charge and from subsequent reactions with oxygen in the surrounding air in the structure. Mixing of the extremely hot explosion gas products with the initial gas in the structure results in an elevated equilibrium temperature of the gas mixture. Since the volume of the structure remains essentially constant during the explosion, the elevated temperature must be accompanied by an increase in the equilibrium pressure of the gas mixture. The process can be viewed to be similar to a combustion test in a bomb calorimeter. The pressure will slowly decay with time due to heat losses to the structure walls; however, in comparison with the highly transient nature of the shock phenomena, this pressure can be truly defined as quasi-static.

Historically in the literature, this quasi-static pressure has been known by different names such as static pressure, steady overpressure, internal blast pressure, post-detonation pressure, and chamber pressure. It is assumed that the reason that no single name has evolved is because there has been misinterpretation of existing names or it has been felt that existing terms do not adequately describe the phenomena. Therefore to add to the growing list of names and hopefully to clarify, this quasi-static pressure created by mixing the hot explosion gas products with the initial gas in the closed structure is simply defined in this report as the confined-explosion gas pressure.

Existing Methods of Calculation. Currently there are two commonly used methods for estimating the magnitude of the confined-explosion gas pressure; that proposed by Filler in references (17) and (18) and that proposed by Weibull in reference (19). Filler proposed that the confined-explosion gas pressure can be calculated from an expression equivalent to

$$\Delta P_g = (4hW)/V_o$$

where ΔP_g = confined-explosion gas pressure (overpressure), psi
 h = heat of combustion of explosive, cal/gm
 W = weight of explosive, lb
 V_o = volume of closed structure, ft³

This method assumes that there is sufficient oxygen in the initial air in the closed structure to ensure that an oxygen-deficient explosive will achieve complete combustion. It also assumes that the specific heat of the gas mixture remains constant. This approach was verified for small quantities of different explosives detonated in a large air-filled chamber resulting in modest confined-explosion gas pressures up to about 30 psi. Realizing the deficiency in the use of the heat of combustion in a possibly oxygen-poor atmosphere, Filler conducted experiments in an inert atmosphere and found results that indicated the heat of detonation yielded accurate agreement for this case, as expected. Unfortunately these studies did not determine analytical relations that describe the phenomena in the transition region between the heat of combustion and heat of detonation. Neither did they extend to the high-pressure region where the effects of variations in gas specific heats could be observed readily.

Weibull proposed that the confined-explosion gas pressure for a TNT charge can be calculated from the expression

$$\Delta P_g = 2410 (W/V_o)^{0.72}$$

This method was an empirical fit to experimentally measured pressures from TNT explosions. Unlike Filler's method, there are no means of relating this equation to explosives other than TNT. However, Weibull's experimental data extends into the high-pressure range (near 1000 psi) where obviously the specific heats of the gas mixture components are changing and the transition between heat of combustion and heat of

detonation can be observed. Unfortunately, this study was limited to an empirical approach without fully exploring the underlying phenomena.

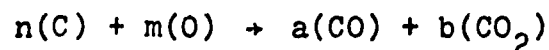
Need for Improved Method. Figure 5.1 gives the prediction curves proposed by Filler's method and Weibull's method. Weibull's extensive TNT data are also plotted for direct comparison. The deficiencies of these two methods become obvious from the comparison. Because complete combustion and a constant specific heat of the gas mixture were assumed, Filler's method becomes decreasingly accurate as the pressure level increases. Even if the heat of detonation is used with Filler's method (the lowest curve in Figure 5.1), the deficiencies in handling the transition region are easily recognized. Weibull's curve approximates the TNT data better than Filler's method over the range of data, but it is all too clear that important physical phenomena are being glossed over in the empirical treatment of the problem that makes it impossible to extend this method to any explosive other than TNT.

Since the confined-explosion gas pressure is believed to be the most important loading parameter in the aircraft internal blast problem, it was imperative that an improved method for calculating the confined-explosion gas pressure be developed. The following sections describe the technique contained in the computer program for predicting the confined-explosion gas pressure and comparing code results with available experimental data.

Description of Improved Method. The improved method assumes an explosion in a closed structure of volume, V_0 , filled with air at some ambient pressure, P_a , and temperature, T_a . The explosive is limited to a hydrocarbon form of the elements C, H, O, and N with aluminum being the only possible metallic additive. Since most explosive compounds are oxygen-deficient, it is assumed that the reaction can consume all of the oxygen in the air in the closed structure, if needed. This basically is assuming optimum mixing and reaction. The code calculates the number of moles of air initially in the closed structure volume from the perfect gas law. One mole of air is assumed

to be composed of 0.21 mole O_2 + 0.79 mole N_2 . From the C, H, O, N, AL composition of the explosive charge given as weight fractions in Table 3.1, the code calculates the number of moles of each of these elements.

The chemical reaction of the explosion and mixing with the air in the closed structure creates the combustion products H_2O , AL_2O_3 , CO, CO_2 , O_2 , and N_2 . A priority in the reaction is assumed as follows; (1) the hydrogen in the explosive reacts with oxygen such that all hydrogen appears as H_2O , (2) the aluminum has next priority on the oxygen, such that all the aluminum appears as the solid AL_2O_3 , (3) if there is an overabundance of oxygen in the explosive and structure air, complete combustion occurs such that all carbon appears as CO_2 and the remaining oxygen not needed in any of the reactions appears as O_2 , (4) if there is insufficient oxygen in the system after the H_2O and AL_2O_3 reactions, then CO and CO_2 are produced in quantities given by the following equations



$$\begin{array}{lcl} a + b = n & \text{or} & a = 2n - m \\ a + 2b = m & & b = m - n \end{array}$$

where

a = number of moles of CO produced
 b = number of moles of CO_2 produced
 n = number of moles of C
 m = number of remaining moles of O

and no O_2 exists in the final gas mixture, (5) the nitrogen does not participate in the reaction and appears as N_2 in the final gas mixture. From the above calculations the number of moles of component gases (H_2O , CO, CO_2 , O_2 , N_2) that make up the final gas mixture in the closed structure are known.

The formation of H_2O , AL_2O_3 , CO, and CO_2 in this combustion-type process releases a large amount of heat energy. Respective standard heats of formation are multiplied by the moles of individual gas components, and the sum of these quantities is defined for use in this report as the heat of reaction. The heats of formation for the gas products are negative by standard thermodynamic terminology, i.e., if energy is released to the surrounding atmosphere, the heat of formation

is negative. Thus the heat of reaction is likewise negative. However, for convenience in this report, it is desirable to express the total amount of energy, Q , released by the explosion as a positive quantity. The heats of formation of the gas products and the heat of reaction are treated as positive quantities in the computer program. To account for the heat of formation of the explosive compound in determining the total energy, Q , it is necessary to add the heat of formation of the explosive compound given in Table 3.1 to the heat of reaction. (Signs of values in Table 3.1 conform to standard thermodynamic terminology.)

As a computational model only, the gas components of the final gas mixture in the closed structure are assumed to exist at the initial ambient pressure, P_a , and temperature, T_a , of the air in the initial volume, V_o . The energy, Q , is then added to the gas mixture, but it is added in 100°F steps in temperature.

It is well known that the addition of heat to a gas in a constant volume system follows the perfect gas relation

$$\Delta Q = n C_v \Delta T \quad (5.1)$$

where

ΔQ = heat added

n = moles of gas

C_v = specific heat of gas at constant volume

ΔT = change in temperature

One of the weaknesses of previous methods for determining the confined-explosion gas pressure was that the variation in C_v with temperature was neglected. Given in the literature, reference (20), are equations relating the specific heat at constant pressure, C_p , with temperature for the various component gases in the final gas mixture. With the assumption that the perfect gas relation

$$R_o = C_p - C_v \quad (R_o = \text{universal gas constant})$$

can be used, equation (5.1) becomes

$$\Delta Q = n (C_p - R_o) \Delta T \quad (5.2)$$

and direct use of the C_p equations in reference (20) can be made. For convenience in calculation, the computer finds a weighted average C_p to be used in equation (5.2) with the total number of moles of gas, n , in the final mixture.

With the total energy released, Q , and the total number of moles, n , of the gas mixture known, the computer uses the following numerical procedure to determine the final temperature of the gas mixture. (The initial temperature is taken at $T = T_a$ and the addition of Q follows a constant volume process.) (1) The weighted average C_p for the gas mixture is determined for the temperature, T . (2) For a temperature step of $\Delta T = 100^\circ\text{F}$, the incremental amount of heat, ΔQ , required to change the temperature by 100°F is calculated from equation (5.2). (3) The temperature of the gas mixture after the step is $T = T + \Delta T$. (4) The incremental energy, ΔQ , is subtracted from the total released energy, Q . (5) The calculational steps (1) through (4) continue until all of the total released energy, Q , is used, thus the final temperature, T_f , is calculated.

With the final temperature, T_f , determined, the perfect gas law gives the final pressure of the gas mixture in the closed volume, V_o , by the relation

$$P_f = n R_o T_f / V_o \quad (5.3)$$

Conventionally this pressure is expressed as an overpressure, so that the confined-explosion gas pressure, P_g , is defined as

$$\Delta P_g = P_f - P_a \quad (5.4)$$

It should be restated that this method of calculating the confined-explosion gas pressure is limited here to C-H-N-O type explosives with aluminum as the only possible metallic additive. Since most common explosives used as fills in conventional weapons fall into this category, this limitation is not considered restrictive to the general use of this computer program. Also this improved method should yield conservative results because optimum mixing and the most efficient chemical reactions are assumed.

Comparison with Experimental Data. Attention is now directed to the adequacy of this improved method. In Figure 5.2 Weibull's TNT data from reference (19) are plotted as indicated by circles. The computer code predictions are given as the solid curve. We note that the agreement with the data is excellent, that the change in slope of the predicted curve follows the general behavior of the data, and that

the data falls either on the curve or slightly below it which demonstrates the conservatism of the new method. From this comparison alone, it is concluded that this technique is far superior to the existing methods of calculating the confined-explosion gas pressure for TNT.

Before assuming the generality of this improved method, it is necessary to make comparisons with experimental data from different explosive mixtures and different initial ambient air conditions. The next most complete set of data is found in reference (21) for a RDX/WAX, 89.5/10.5 mixture detonated in air at sea level conditions. A plot of the data points (circles) from this study and the code predictions (solid curve) are given in Figure 5.3. Again the excellent agreement and the conservatism of the improved method predictions are noted. Reference (21) also gives data for this same RDX/WAX mixture for a reduced atmosphere ($P_a = 1$ psia). These data and the code predictions are given in Figure 5.4, and the same excellent agreement and conservatism are demonstrated.

Other assorted data for different explosives were found in reference (21), and some aluminized explosive data were found in reference (17)--all for sea level conditions. There was an insufficient quantity of these data to construct curves, thus they are tabulated in Table 5.1 along with calculated code predictions. The excellent agreement is again noted, especially for the extremely high-pressure PETN data and the aluminized RDX data. An interesting observation can be made with the aluminized data. As the percentage of aluminum increases, the overprediction of the confined-explosion gas pressure tends to increase. But even for the unrealistic mixture containing 50% aluminum, there is only a 16% deviation. This increase is believed due to the assumed optimum mixing and most efficient reaction in the code. Evidently the aluminum is not able to utilize the oxygen in the surrounding air to the maximum extent assumed in the code calculation.

It is concluded from these comparisons that the improved method for calculating the confined-explosion gas pressure is far superior to any other known existing technique. Even with the use of perhaps a not-too-realistic combustion type model and the liberal use of

equilibrium perfect gas relations and properties for high-pressure and temperature transient conditions, the improved method appears to perform exceptionally well. From these comparisons the method appears capable of handling mono, composite, and aluminized explosives at sea level ambient conditions and reduced atmospheric ambient conditions. Therefore with justifiable confidence, this improved method is used as the basis for confined-explosion gas pressure calculations in the computer program.

Although use of this computer program has been consistently limited to C-H-N-O explosives with aluminum as the only possible metallic additive, there is no theoretical reason why it cannot be adjusted to perform well with other metallic additives or non-C-H-N-O explosives. This limitation arises only because there exists no experimental data on confined-explosion gas pressure for these different explosives that will permit the establishment of a set of reaction priorities similar to those for C-H-N-O explosives.

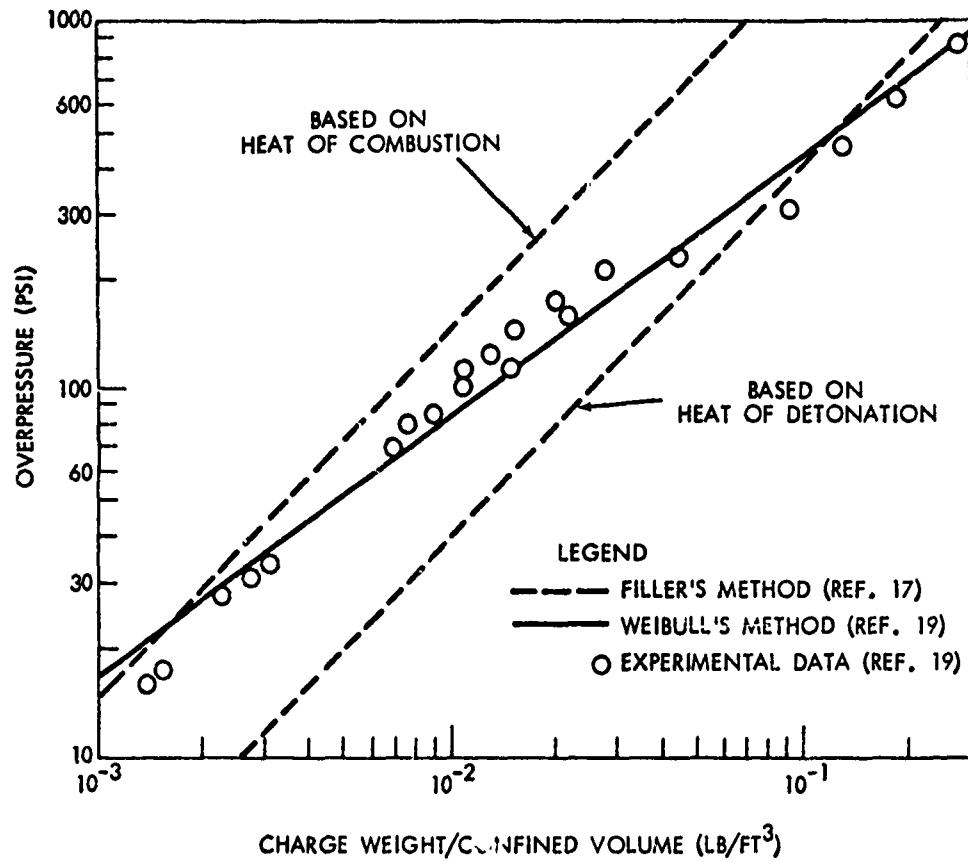


FIG. 5.1 COMPARISON OF EXISTING METHODS FOR PREDICTING CONFINED-EXPLOSION GAS PRESSURE WITH EXPERIMENTAL DATA FOR TNT EXPLOSIONS IN AIR AT SEA LEVEL CONDITIONS

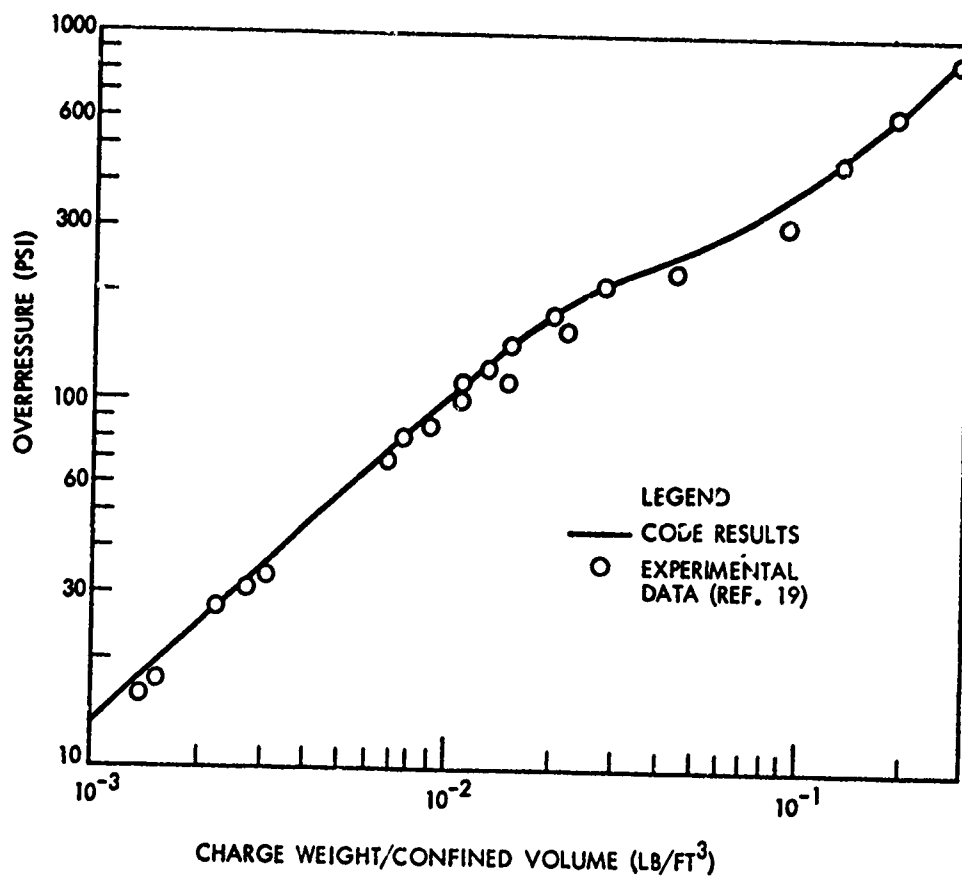


FIG. 5.2 CODE RESULTS FOR CONFINED-EXPLOSION GAS PRESSURE FOR TNT IN AIR AT SEA LEVEL

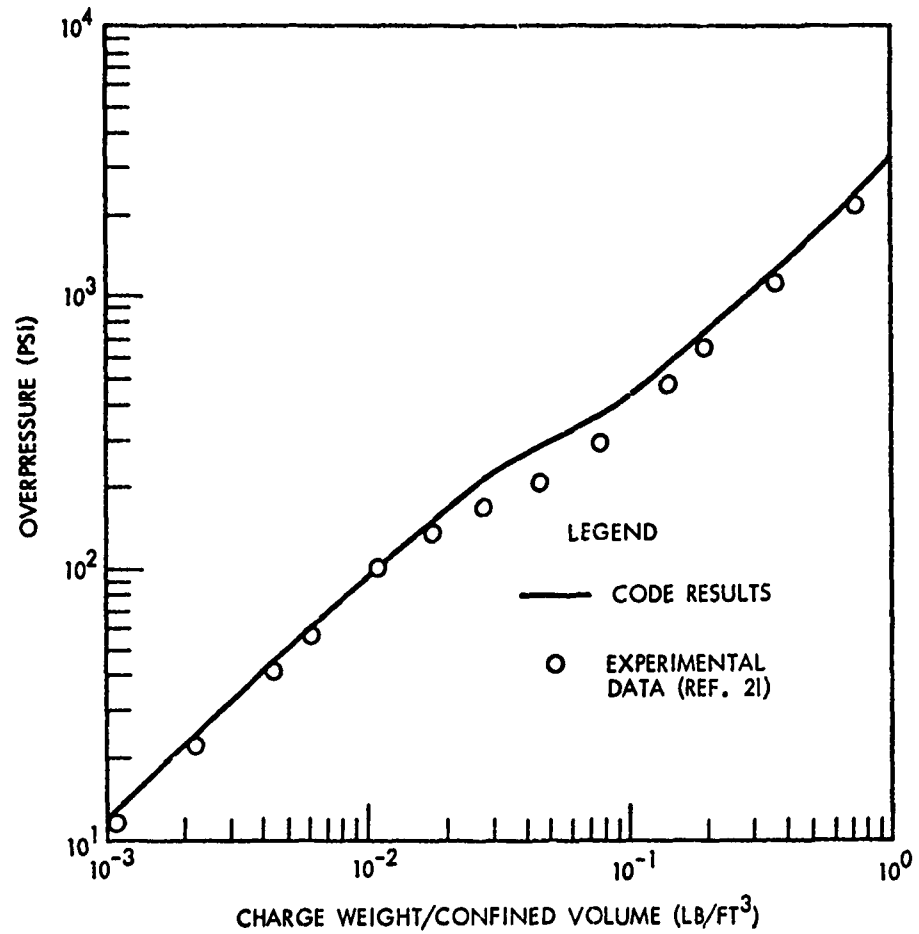


FIG. 5.3 CODE RESULTS FOR CONFINED-EXPLOSION GAS PRESSURE FOR RDX/WAX, 89.5/10.5 IN AIR AT SEA LEVEL

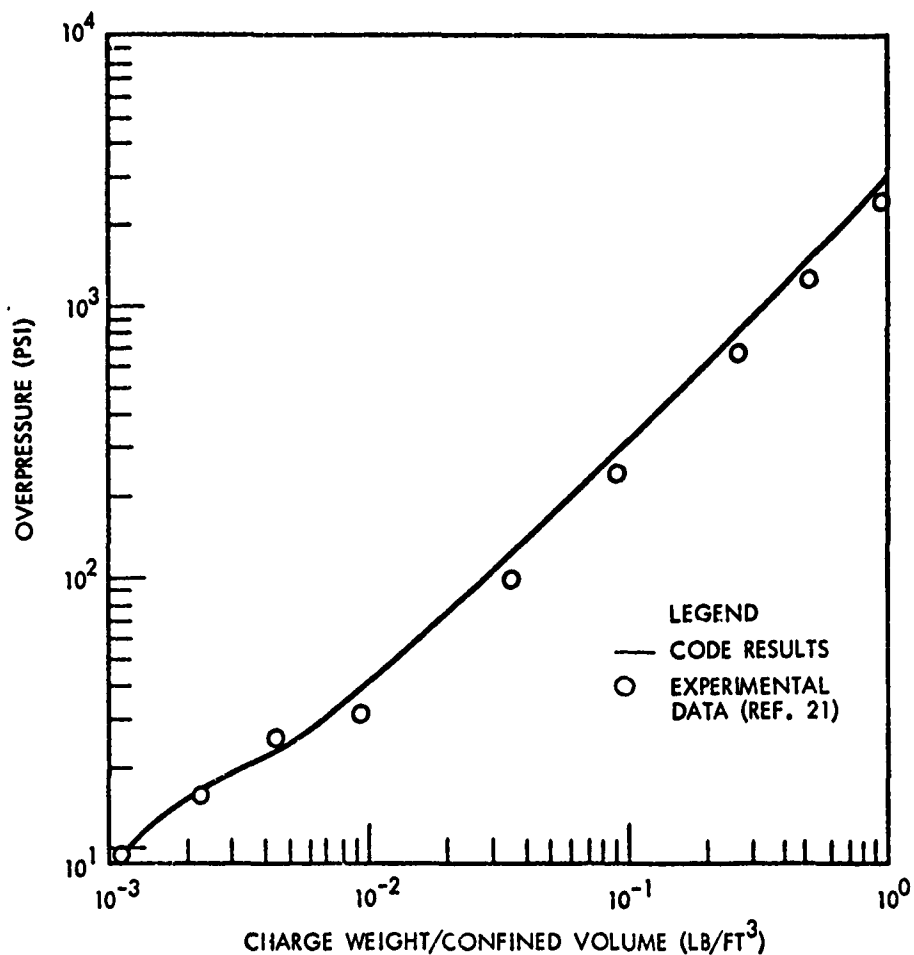


FIG. 5.4 CODE RESULTS FOR CONFINED-EXPLOSION GAS PRESSURE FOR RDX/WAX, 89.5/10.5 IN AIR AT P = 1 PSI

TABLE 5.1
MISCELLANEOUS CONFINED-EXPLOSION
GAS PRESSURE DATA

<u>TYPE OF EXPLOSIVE</u>	<u>W/V (LB/FT³)</u>	<u>CALCULATED OVERPRESSURE (PSI)</u>	<u>EXPERIMENTAL OVERPRESSURE (PSI)</u>	<u>DEVIATION (%)</u>
RDX/TNT				
60/40	0.00221	22.0	19.9	+10
60/40	0.00442	41.0	38.3	+ 7
PETN				
PETN	0.182	711	725	- 2
PETN	0.304	1089	1110	- 2
PETN	0.405	1405	1400	+ 0
RDX/AL/WAX				
98/ 0/2	0.00171	15.3	15.6	- 2
76/22/2	0.00171	22.0	21.3	+ 3
63/35/2	0.00171	26.6	24.3	+ 9
48/50/2	0.00171	30.3	26.0	+16

DATA FROM REFS. 17 AND 21

CHAPTER 6

VENTING CALCULATIONS

Venting. Inherent in the preceding section was the assumption of a completely closed structure, i.e., no venting occurs before the maximum value of the confined-explosion gas pressure is established. Therefore, at the onset of venting the initial conditions of the confined-explosion gas pressure are known from previous calculations; P_f (gas pressure in absolute units), T_f (temperature), n (total number of moles of gas), C_p (average specific heat of gas at T_f), and V_o (volume of gas). A combination of n , V_o , and the molecular weights of the gas mixture components yields the initial density, ρ_f , of the gas mixture. Gamma, γ , (the ratio of specific heats), is found with the known value of C_p from the perfect gas relation

$$\gamma = C_p/C_v = C_p/(R_o - C_p) \quad (6.1)$$

From code input information, the constant backpressure, P_b , against which venting occurs and the initial vent area, A_o , are given.

The relations governing the venting process have been derived in reference (22) for steady isentropic flow through a perfect nozzle. In this reference, γ was taken to be 1.4 which permitted the relations to be expressed in closed form. However, since γ in this computer program is not 1.4 and is not constant, the differential form of these governing equations are taken from reference (22). (There is a typographical error in equation (15) of reference (22)--($\gamma - 1$) in the denominator should be ($\gamma + 1$).) These governing equations are:

for sonic flow

$$\frac{\Delta P_1}{P_1} \frac{3\gamma-1}{2\gamma} = \left[g \gamma^3 (P_o^{1/\gamma}/\rho_o) \left(\frac{2}{\gamma+1} \right)^{\frac{\gamma+1}{\gamma-1}} \right]^{1/2} \frac{A_o}{V_o} \Delta t \quad (6.2)$$

$$\text{for } P_1 \geq \frac{P_b}{\left(\frac{2}{\gamma+1}\right)^{\frac{\gamma}{\gamma-1}}}$$

for subsonic flow

$$\frac{P_1^{\frac{\gamma-1}{\gamma}} \Delta P_1}{\left(P_1^{\frac{\gamma-1}{\gamma}} - P_b^{\frac{\gamma-1}{\gamma}}\right)^{1/2}} = \left[g \gamma^3 \left(\frac{2}{\gamma-1}\right) \left(\frac{P_o}{\rho_o} \frac{P_b^2}{\gamma}\right)^{1/\gamma} \right]^{1/2} \frac{A_o}{V_o} \Delta t \quad (6.3)$$

$$\text{for } P_b < P_1 < \frac{P_b}{\left(\frac{2}{\gamma+1}\right)^{\frac{\gamma}{\gamma-1}}}$$

Throughout the venting process, these isentropic relations are assumed

$$T_1 = T_o (P_1/P_o)^{(\gamma-1)/\gamma} \quad (6.4)$$

$$\rho_1 = \rho_o (P_1/P_o)^{1/\gamma} \quad (6.5)$$

The terms are defined as follows

$\Delta P_1 = (P_o - P_1)$ = arbitrary pressure step increment

P_o = pressure at beginning of increment step

P_1 = pressure at end of increment step

P_b = ambient backpressure (constant throughout venting)

g = acceleration due to gravity

γ = specific heat ratio given by equation (6.1) (based on T_o and assumed constant during increment step)

ρ_o = density at beginning of increment step

ρ_1 = density at end of increment step

T_o = temperature at beginning of increment step

T_1 = temperature at end of increment step

V_0 = volume of structure (constant)

A_0 = vent area (constant)

Δt = time increment (parameter to be determined)

Even though the relations are derived for steady flow, it is assumed that they are applicable to the venting problem because the pressure step used in the numerical solution of these equations is sufficiently small that gas mixture properties can be considered constant during a single incremental step.

It is assumed in the venting process that the composition of the gas mixture remains constant, i.e., no single component of the gas mixture is vented preferentially. The pressure increment step used in the computer program is defined as

$$\Delta P_1 = (P_f - P_b)/100 \quad (6.6)$$

i.e., there are 100 pressure increment steps. Even though the computer performs 100 steps in this calculation, only every tenth step is printed as output. The printout can be increased to up to every other step if desired. During the venting process as the gas density is decreasing, the computer keeps a running account of the quantities of each component of the gas mixture remaining in the compartment structure. The need for this procedure will become apparent in subsequent sections.

The following discussion is a description of a typical venting calculation. (1) Starting values of P_0 , ρ_0 , T_0 , γ , A_0 , V_0 , and P_b are known. (2) From the pressure increment step, P_1 is calculated and used to determine if flow is sonic or subsonic. (3) With the proper equation (6.2) or (6.3) chosen, the time increment Δt is determined, and from $t_1 = t_0 + \Delta t$ the absolute time from beginning of venting associated with P_1 is found. (4) From equations (6.4) and (6.5) the temperature and density at the end of the increment step are determined. (5) From the density the number of moles of gas mixture components remaining in the compartment are found. (6) From the temperature, a new average C_p is calculated from which a new γ is determined. (7) Values at the end of this increment step, P_1 , T_1 , ρ_1 , and new γ ,

become the beginning values P_o , T_o , ρ_o , and γ for the next step.

(8) The above procedures form a loop that continues until the 99th increment step is completed which is the step immediately before $P_1 = P_b$. The program is stopped here because equation (6.3) cannot be solved for $P_1 = P_b$.

Vent Area and Volume Changes. The orderly procedure given above can be readily interrupted to accommodate vent area and volume changes in accordance with input failure criteria controlling damage propagation. By constantly monitoring the pressure-time history of the venting process, the computer can easily adjust to changes from input of the type presented in Figure 3.1. For a given pressure or time, an adjustment in vent area can be made simply by changing the value A_o in equations (6.2) and (6.3). However, a volume change requires not only the change of V_o in equations (6.2) and (6.3) but also an adjustment in the gas mixture pressure because the volume has changed.

Upon wall failure in the initial compartment, it is assumed that the gas mixture in the initial compartment instantaneously mixes and comes to equilibrium with the air in the newly available compartment. The conservation of energy states for this process that the sum of the internal energy of the gas mixture immediately prior to wall failure and the internal energy of the air contained in the adjacent compartment is equal to the internal energy of the new gas mixture after the mixing process. (No further chemical reaction is assumed to occur.) In equation form, this concept is stated as

$$\frac{P_1 V_o}{(\gamma_1 - 1)} + \frac{P_a V_a}{(\gamma_a - 1)} = \frac{P_2 (V_o + V_a)}{(\gamma_2 - 1)} \quad (6.7)$$

where P_1 = pressure of gas mixture immediately prior to wall failure

γ_1 = gamma of gas mixture immediately prior to wall failure

V_o = volume of gas mixture immediately prior to wall failure

P_a = ambient pressure of air in adjacent compartment

V_a = volume of air in adjacent compartment

γ_a = gamma of air (taken to be 1.4)

P_2 = pressure of new gas mixture

γ_2 = gamma of new gas mixture

Unfortunately there are two unknowns in equation (6.7), P_2 and γ_2 . (γ_2 is unknown because the gas composition and temperature have changed.)

By keeping a running account of the amounts of the components of the gas mixture before wall failure and by calculating the amount of oxygen and nitrogen in the air in the new compartment, the computer calculates the composition of the new gas mixture and finds the total number of moles of the new gas mixture, n_2 . Since the new volume ($V_o + V_a$) is known, the new density, ρ_2 , is calculated. The perfect gas law

$$P_2 = (n_2 R_o T_2) / (V_o + V_a) \quad (6.8)$$

gives a second relation but introduces the third variable, T_2 . From the programmed C_p equations as a function of temperature for the gas components, the computer is capable of generating a third relation from the known quantities of the gas components in the new mixture

$$\gamma_2 = \gamma_2(T_2) \text{ (based on equation (6.1))} \quad (6.9)$$

The numerical iteration solution of equations (6.7)--(6.9) gives the values of P_2 , T_2 , and γ_2 . Since ρ_2 , n_2 , and the new gas mixture components are known, all of these values become the beginning parameters for the next increment step in the venting calculation method. Subsequent wall failures controlled by input damage criteria are treated in this same manner.

Verification. There are no experimental data available to verify this entire venting process including vent area and volume changes. There are only limited data applicable to the venting process without area and volume changes. These are given in reference (22) from which the venting equations were taken. Here venting of the confined-explosion gas pressure from a test facility at NOL was measured. Venting gases escaped the test chamber of the facility through a

"S" shaped labyrinth passageway out either a partially open door (small vent area) or an open door (medium vent area). Agreement between equation predictions and experimental data was very good for the small vent area where flow in the passageway was probably sufficiently slow not to induce any type of flow losses. Agreement for the medium vent area, which was about seven times greater than the small vent area, was only fair with the equations underpredicting vent times by 20 to 30%. It is believed that the seven-fold increase in vent area produced relatively high flow velocities in the passageway from which significant losses slowed the venting process.

In reference (22) caution is expressed in using these venting relations for large vent areas. However, it has been learned that limited unpublished data* from the Naval Ship Research and Development Center (NSRDC) on venting explosion gases through large openings agree very well with predictions from the venting equations. Therefore, with only limited confirmation of the venting procedure, this method is employed in the computer code for predicting the pressure-time history of the confined gas mixture.

Limitations. The venting section of the computer code has not been verified experimentally to any significant degree. Thus experimental evidence in this area is needed to assign a confidence level to this section equivalent to that of the shock and confined-explosion gas pressure sections. Four assumptions are made in this section that need additional study. First, heat losses to the surrounding walls are neglected as a significant mechanism to reduce the gas mixture pressure. Second, a constant backpressure against which venting must occur is assumed. Third, gas mixing and the establishment of pressure equilibrium occur instantaneously with compartment wall failure. Fourth, no chemical reactions occur with the air in the adjacent compartments after wall failures. In terms of the small compartments in aircraft wings and significant venting to the atmosphere, none of these assumptions are believed to be restrictive for aircraft applications. However, for large explosions and large structures such as ship

* Information received from J. W. Sykes (NSRDC).

compartments or building rooms, they may indeed be restrictive and may require additional study and modification. The code is constructed in a manner such that modifications in these areas can be easily accommodated.

CHAPTER 7

USER'S GUIDE

Computer Requirements. The computer program is written in FORTRAN for a CDC 6400 computer, and it should work without change on other CDC machines. The program is straightforward and can be adapted easily to other computers. The major change that may have to be made is the spreading out onto individual cards of the statements that are now placed on a single card and separated by the \$ sign. Storage requires less than 32,000 core memory words, the compilation time is about 15 seconds, and the run time for a single case is about 1 second on the CDC 6400.

Program Structure. A complete flow diagram of the computer program is given in Appendix A. Detail descriptions of the input cards and format are given in Appendix B. A complete list of the program variables with their definitions are given in Appendix C. The code consists of the main program BLAST and six subroutines. The functions of these sections are as follows:

- BLAST: reads input data; does venting calculation; does final portion of the shock wave calculation
- MIX: supplies new conditions (pressure, volume, temperature, gamma) after the gases of two compartments are mixed.
- HEDATA: contains tables of properties of explosive components and mixtures.
- GAMMA: supplies average specific heat ratio and internal energy for a gas of given composition and temperature.
- GASES: supplies initial conditions in the compartment immediately after the explosion occurs and the confined-explosion gas pressure is developed.

TNT: supplies pressure, distance, arrival time, and other data for a spherical 1-lb TNT free-field explosion at sea level.

ARDC: gives standard-atmosphere pressure and temperature for a desired altitude.

A complete listing of the entire program is given in Appendix D.

Printed Warnings in Output. During the running of a problem with this computer program, printed diagnostic or warning statements may appear in the output to alert the user. These are:

- (1) "WFACT NOT KNOWN, 1.0 IS USED."

This means that no energy equivalent weight has been supplied for the shock wave calculation with the desired explosive. The program assumes a value of 1.0, thus results are equal to those of TNT.

- (2) "CHARGE SHAPE CORRECTION IS CRUDE. PSI EXCEEDS RANGE OF EXPERIMENTAL DATA."

The cylindrical charge equivalent weight depends on the peak shock pressure level. Above 100 psi, no experimental data were available for correlation and theoretical techniques were used. The warning statement is printed to indicate that shock data for the particular case under study are approximations.

- (3) "CASE WEIGHT CORRECTION IS CRUDE. PSI EXCEEDS RANGE OF EXPERIMENTAL DATA."

The method for calculating the casing equivalent weight was based on pressure data for 100 psi and below. The warning statement is printed to indicate that shock data for the particular case under study are approximations because the peak overpressure exceeds 100 psi.

- (4) "CAUTION--CONTACT SURFACE HAS ARRIVED. DATA ARE CRUDE BEYOND T(MSEC) AFTER SHOCK ARRIVAL =."

This warning statement appears during the shock calculations if the contact surface reaches the desired distance being investigated. It indicates that, after the indicated time, the shock data are approximations.

Changes to the Program. There are several items in the computer program that probably will be frequently changed by the user depending upon the problem under consideration. They involve the addition of new explosives to the data table in the program and changes in the amount of printout for the shock and venting calculations. If a new explosive not in the data bank is frequently used, the user may wish to add it permanently to the subroutine HEDATA. There is room for 11 new explosives in this table. Beginning with index number 30 (card HEDA0675), four data cards using the format for the existing explosives can be inserted to input the new explosive.

The amount of printout desired for shock and venting calculations may vary with the area of interest for a particular problem. This is easily changed by varying KMAX1 for shock calculations or KMAX2 for venting calculations on card BLAS0560. The shock wave calculation is done in KMAX1 steps in time, equally spaced within the positive duration of the overpressure. The built-in value of KMAX1 is 10. If more printout is desired, change KMAX1 to 20 or 40. If more than 40 lines are desired, the dimensions of PSI(40), T1(40), T2(40), and PSIREF(40) must be increased. Values of KMAX1 below 10 are not recommended because the numerical integration to obtain impulse is controlled by the number of these steps.

The venting calculations are performed in 100 fixed integration steps. However, only every KMAX2-th step is printed as output. The built-in value of KMAX2 is 10, giving 10 lines of venting printout. If more venting data is desired, KMAX2 can be changed to 5, 4, or 2.

Example Problems. To demonstrate the use of the computer code with its different options and features, nine sample problems have been run. They are variations of the following base problem.

Consider an 8 ft³ compartment. A projectile has penetrated the compartment forming an opening of 0.00545 ft² area. The projectile contains 0.0294 lb of explosive of composition 74% RDX, 21% AL, and 5% WAX. It has a length to diameter ratio of 2.7 and a case weight to charge weight ratio of 4.24.

Figure 7.1 shows the input cards for the nine problems that have been solved. (Appendix B gives the description and format of input data cards.) Examples 1-3 show the three ways to specify the explosive.

Examples 5-7 show the three options specifying damage criteria for wall failure in the venting calculations. Example 8 shows the method of a shock wave calculation with several distances specified. Examples 4 and 9 demonstrate problems at an altitude other than sea level. Specific descriptions of each example are given in the following paragraphs.

Example 1 involves only one input card. The explosive number is 17, indicating the number of the desired explosive in the list of subroutine HEDATA. At the end of the card, NOPT=1 indicates that only a venting calculation is desired; NV=0 indicates that the chamber volume and vent area remain at their initial values throughout the problem; and NR=0 indicates that no distances are specified since this is only a venting calculation. The results of this problem are shown in Figure 7.2.

Example 2 involves two data cards. The first card is the same as in Example 1 except that the explosive number is 0. This causes the second card to be read in. This card gives the energy equivalent weight = 1.30, the heat of formation = 29.36 cal/gm, and the weight fractions of C, H, O, N, and AL. The results are the same as for Example 1 which are given in Figure 7.2.

Example 3 again involves two data cards. The first card has an explosive number of -1. This causes the second card to be read in. This card gives the energy equivalent weight = 1.30 and the weight fractions of the desired components from the explosive list in HEDATA: 74% number 27 (RDX), 21% number 25 (AL), and 5% number 26 (WAX). Again, the results are the same as those of Example 1 which are given in Figure 7.2.

Example 4 is the same as Example 1 except that the compartment is at altitude rather than at sea level. The ambient pressure is 6.76 psia and the temperature is -24.6°C. The results of this calculation are shown in Figure 7.3.

Example 5 returns to sea level but the compartment volume is allowed to change. NV=1 means that one card of volume and area change data is to be read. This card contains the following data; if the confined-explosion gas pressure in the tank exceeds 30 psia, the volume increases by 4 ft³ and the vent area increases 0.00545 ft².

The ambient pressure and temperature of the air in this additional volume and vent area are added only if the confined-explosion gas pressure exceeds the 45-psia level. The results are shown in Figure 7.4.

Example 6 has two cards of volume and area change data. The last number on these cards is 2 which indicates that the changes of volume and area are to be made at the indicated times of 0.15 and 0.60 sec. No tests are made on the pressure, so that the changes are made at the indicated times regardless of the pressure. The results are shown in Figure 7.5.

Example 7 has three cards of volume and area data. The last number on these cards is 3 which indicates that if the pressure exceeds the indicated value when the indicated time is reached, then the volume and area change is made. For example, if the pressure in the tank exceeds 45 psia at 0.15 sec, 4 ft³ of volume and 0.00545 ft² of vent area are added. The results are shown in Figure 7.6.

Example 8 is a shock wave calculation only, indicated by NOPT=2. NV=0 since no venting parameters are involved in a shock wave calculation, and NR=3 since three distances are desired. The second card contains these three distances: 0.667, 1.000, and 1.333 feet from the center of the charge. The results for the single distance of 0.667 are shown in Figure 7.7.

Example 9 is the same as Example 8 except that the compartment is at altitude. The results are shown in Figure 7.8.

Explanation of Typical Output. A typical example of the printout for shock calculations is given in Figure 7.7 which are the results from Example 8. The index number and properties of the explosive used in the calculation appear at the beginning of the output. The two warning statements concerning the cylindrical charge equivalent weight and casing equivalent weight are noted. Under "SHOCK WAVE CALCULATION", the left-hand column repeats all the input parameters governing the shock problem. In the right-hand column, certain constants derived by the computer for the calculation are given:

ADJUSTED WT(LB TNT)--equivalent TNT sphere from equation (4.10)
HE ENERGY FACTOR--energy equivalent weight, f_e , from Table 3.1

CHARGE WEIGHT FACTOR	cylindrical charge equivalent weight, f_s , from Figure 4.5
CASE WEIGHT FACTOR	casing equivalent weight, f_c , from equation (4.8) or (4.9)
PRESSURE SCALE FACTOR	(P_s/P_a) for equations (4.2)--(4.5)
DISTANCE SCALE FACTOR	$(W_s/W_a)^{1/3} (P_a/P_s)^{1/3}$ for equations (4.2)--(4.5)
TIME SCALE FACTOR	$(W_s/W_a)^{1/3} (P_a/P_s)^{1/3} (T_a/T_s)^{1/2}$ for equations (4.2)--(4.5)
NORMAL REFL FACTOR	normal reflection factor, f_R , from equations (4.11)--(4.13)

The tabulated pressure-time shock data is noted for the desired distance of 0.667 ft. Both the incident and normally reflected overpressures are given as functions of time where time is measured from the instant of detonation and shock arrival. For example, the shock arrives at the distance 0.667 ft in 0.07118 msec; the peak incident overpressure is 317.6 psi and the reflected overpressure is 2144 psi; and the positive phase of the shock is completed 0.192 msec after detonation or 0.1209 msec after shock arrival. Next the impulses for the incident and reflected waves are given. Lastly, the warning statement concerning the contact surface appears which states that 0.02787 msec after the shock arrives, the pressure-time data are approximations.

Figure 7.6 gives printout results for Example 7 on the confined-explosion gas pressure venting and subsequent changes due to structural failures. At the beginning of the output are the index number and properties of the explosive used in the calculation. Under "VENTING CALCULATION" a repeat of input parameters is given; under "BEGIN VENTING CALCULATION" the input failure criteria table is repeated. Under "PROPERTIES OF GASES" the output describes the condition of the confined-explosion gas in the initial compartment volume before any venting has occurred. The first statement indicates that oxidation was complete, i.e., sufficient oxygen was available to make H_2O , Al_2O_3 , and only CO_2 . Had there been insufficient oxygen for complete oxidation, the output would have indicated the name and quantity of the last product formed. For example, if all $H \rightarrow H_2O$

and $AL \rightarrow AL_2O_3$ but there was insufficient oxygen to completely react with all the C to form CO, the computer would print "PERCENT LAST PRODUCT (CO) = (fraction of carbon used)". Next, the computer prints the maximum temperature of the confined-explosion gas, the energy released in the chemical reaction that creates the confined-explosion gas pressure, the specific heat ratio, γ , and the maximum value of the confined-explosion gas pressure expressed as an overpressure.

Under "BEGIN VENTING OF GASES" the gas pressure-time data are tabulated along with the amount of gas in the confining volume (GASES), the temperature of the gas (TEMP), the specific heat ratio (GAMMA), and an index (NEQN). If this index is 1 then the flow velocity is sonic; if 2, flow velocity is subsonic. The beginning time is zero for this calculation which is set arbitrarily after the dissipation of the shock wave, and the overpressure is maximum at 45.9 psi. Adjustments made with compartment failures and continued venting are noted. For example, at $t=0.15$ sec the gas overpressure is 36.5 psi which is above 45 psia and the wall fails. A new pressure of 36.75 psia or 22 psi overpressure is calculated for the new volume of 12 ft^3 , and venting continues through the new area of 0.0109 ft^2 until $t=0.6$ sec when another failure occurs. The code readjusts the pressure to accommodate the new volume and venting continues until the overpressure is essentially zero at $t=0.9$ sec.

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ITEMS ON FIRST DATA CARD		EXPLOSIVE WEIGHT	EXPLOSIVE NUMBER	L/D	M/C	V ₀	A ₀	P ₀	T ₀	ALTITUDE	P _a	T _a	NOPT	NV	NR
EXAMPLE 1		.0294	17	2.7	4.24	8.	545-3	14.7	20.	0.	14.7	20.	1	0	0
EXAMPLE 2		.0294	0	2.7	4.24	8.	545-3	14.7	20.	0.	14.7	20.	1	0	0
		1.30	29.36	.163	.027	.280	.320	.210							
EXAMPLE 3		.0294	-1	2.7	4.24	8.	545-3	14.7	20.	0.	14.7	20.	1	0	0
		1.30	27	.74	25	.21	26	.05							
EXAMPLE 4		.0294	17	2.7	4.24	8.	545-3	6.76-24.6		0.	6.76-24.6		1	0	0
EXAMPLE 5		.0294	17	2.7	4.24	8.	545-3	14.7	20.	0.	14.7	20.	1	1	0
		30.	0.	4.	.545-2	14.7	20.	1							
EXAMPLE 6		.0294	17	2.7	4.24	8.	545-3	14.7	20.	0.	14.7	20.	1	2	0
		0.	.15	4.	.545-2	14.7	20.	2							
		0.	.60	4.	.545-2	14.7	20.	2							
EXAMPLE 7		.0294	17	2.7	4.24	8.	545-3	14.7	20.	0.	14.7	20.	1	3	0
		45.	.15	4.	.545-2	14.7	20.	3							
		20.	.6	4.	.545-2	14.7	20.	3							
		19.	.8	4.	0.	14.7	20.	3							
EXAMPLE 8		.0294	17	2.7	4.24	8.	545-3	14.7	20.	0.	14.7	20.	2	0	3
		.667	1.0	1.333											
EXAMPLE 9		.0294	17	2.7	4.24	8.	545-3	6.76-24.6		0.	6.76-24.6		2	0	3
		.667	1.0	1.333											

FIG. 7.1 INPUT CARDS FOR NINE EXAMPLE PROBLEMS

INTERNAL BLAST DAMAGE MECHANISMS PROGRAM, MAR 1972
RDX/AL/WAX: 74/21/5

EXPLOSIVE PROPERTIES

NUMBER EQWT EFORM EXPLOSIVE COMPOSITION BY WEIGHT

	KCAL/G	C	H	N	O	AL
17	1,300	.029360	.163	.027	.280	.320
						.210

VENTING CALCULATION

CHARGE WEIGHT(LB) = .2940E-01
 INIT VOLUME(CU FT) = 8.000
 INIT VENT AREA(SQ FT) = .5450E-02
 AMBIENT PRESSURE(PSIA) = 14.70
 AMBIENT TEMP(C) = 20.00
 CHAMBER PRESSURE(PSIA) = 14.70
 CHAMBER TEMP(C) = 20.00
 NOPT= 1 NV= 0

BEGIN VENTING CALCULATION

PROPERTIES OF GASES==

OXIDATION COMPLETE
 TEMPERATURE, DEGREES F = 1653.2
 ENERGY RELEASE(KCAL/G) = 3.4573
 SPECIFIC HEAT RATIO = 1.3141
 GAS OVERPRESSURE(PSI) = 45.945

BEGIN VENTING OF GASES

OVERPR(PSI)	TIME(SEC)	GASES(LB)	TEMP(R)	GAMMA	NEON
45.94	0.	.6167	2113.	1.3141	
41.35	.6936E-01	.5809	2074.	1.3155	1
36.76	.1453	.5443	2032.	1.3167	1
32.16	.2293	.5070	1986.	1.3181	1
27.57	.3229	.4689	1937.	1.3197	1
22.97	.4285	.4297	1884.	1.3214	1
18.38	.5496	.3895	1825.	1.3235	1
13.78	.6911	.3479	1759.	1.3259	1
12.47	.7360	.3358	1739.	1.3267	1
7.880	.9175	.2921	1661.	1.3298	2
3.286	1.171	.2462	1570.	1.3338	2
.6943E-01	1.690	.2125	1494.	1.3380	2

FIG. 7.2 OUTPUT RESULTS FOR EXAMPLES 1, 2, AND 3

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INTERNAL BLAST DAMAGE MECHANISMS PROGRAM, MAR 1972
RDX/AL/WAX, 74/21/5

EXPLOSIVE PROPERTIES

NUMBER EQWT EFORM EXPLOSIVE COMPOSITION BY WEIGHT

	KCAL/G	C	H	N	O	AL
17	1.300	.029360	.163	.027	.280	.320
					.320	.210

VENTING CALCULATION

CHARGE WEIGHT(LB) = .2940E-01
INIT VOLUME(CU FT) = 8.000
INIT VENT AREA(SQ FT) = .5450E-02
AMBIENT PRESSURE(Psia) = 6.760
AMBIENT TEMP(C) = -24.60
CHAMBER PRESSURE(Psia) = 6.760
CHAMBER TEMP(C) = -24.60
NOPT= 1 NV= 0

BEGIN VENTING CALCULATION

PROPERTIES OF GASES--

OXIDATION COMPLETE
TEMPERATURE, DEGREES F = 2655.7
ENERGY RELEASE(KCAL/G) = 3.6573
SPECIFIC HEAT RATIO = 1.2893
GAS OVERPRESSURE(Psi) = 42.914

BEGIN VENTING OF GASES

OVERPR(Psi)	TIME(SEC)	GASES(LB)	TEMP(R)	GAMMA	NEQN
42.91	0.	.3426	3116.	1.2893	
38.62	.6729E-01	.3194	3053.	1.2905	1
34.33	.1420	.2958	2985.	1.2915	1
30.04	.2260	.2716	2912.	1.2926	1
25.75	.3215	.2468	2831.	1.2939	1
21.46	.4323	.2212	2741.	1.2953	1
17.17	.5635	.1968	2640.	1.2971	1
12.87	.7241	.1673	2523.	1.2993	1
8.583	.9296	.1384	2383.	1.3022	1
5.647	1.111	.1176	2268.	1.3049	1
1.356	1.524	.8499E-01	2052.	1.3104	2
.6846E-01	1.968	.7451E-01	1970.	1.3137	2

FIG. 7.3 OUTPUT RESULTS FOR EXAMPLE 4

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INTERNAL BLAST DAMAGE MECHANISMS PROGRAM, MAR 1972
RDX/AL/WAX: 74/21/5

EXPLOSIVE PROPERTIES

NUMMER EQWT EFORM EXPLOSIVE COMPOSITION BY WEIGHT

	KCAL/G	C	H	N	O	AL
17	1,300	.029360	.163	.027	.280	.320 .210

VENTING CALCULATION

CHARGE WEIGHT(LB) = .2940E-01
INIT VOLUME(CU FT) = 8.000
INIT VENT AREA(SQ FT) = .5450E-02
AMBIENT PRESSURE(PSIA) = 14.70
AMBIENT TEMP(C) = 20.00
CHAMBER PRESSURE(PSIA) = 14.70
CHAMBER TEMP(C) = 20.00
NOPT = 1 NV = 1

BEGIN VENTING CALCULATION

TABLE OF VOLUME AND VENT AREA CHANGES

P(PSIA)	T(SEC)	V(CU FT)	A(SQ FT)	PAMB(PSIA)	TAMB(C)	NOPTV
30.00	0.	4.000	.5450E-02	14.70	20.00	1

PROPERTIES OF GASES--

OXIDATION COMPLETE
TEMPERATURE, DEGREES F = 1653.2
ENERGY RELEASE(KCAL/G) = 3.6573
SPECIFIC HEAT RATIO = 1.3141
GAS OVERPRESSURE(PSI) = 45.945

FAILURE LEVEL IN TABLE EXCEEDED.

VOLUME INCREASE(CU FT) = 4.000
NEW TOT VOL (CU FT) = 12.00
NEW TOT AREA (SQ FT) = .1090E-01
NEW PRESSURE(PSIA) = 43.06
NEW GAMMA = 1.338

BEGIN VENTING OF GASES

OVERPR(PSI)	TIME(SEC)	GASES(LB)	TEMP(R)	GAMMA NEQN
28.36	0.	.9162	1500.	1.3377
25.52	.5174E-01	.8708	1475.	1.3409 1
22.69	.1077	.8246	1447.	1.3424 1
19.85	.1616	.7776	1419.	1.3439 1
17.01	.2353	.7296	1388.	1.3456 1
14.18	.3090	.6806	1355.	1.3475 1
12.67	.3515	.6541	1336.	1.3488 1
9.835	.4394	.6032	1299.	1.3509 2
7.000	.5419	.5508	1258.	1.3535 2
4.164	.6696	.4967	1212.	1.3565 2
1.329	.8619	.4406	1161.	1.3601 2
.1943	1.037	.4175	1139.	1.3622 2

FIG. 7.4 OUTPUT RESULTS FOR EXAMPLE 5

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INTERNAL BLAST DAMAGE MECHANISMS PROGRAM, MAR 1972
RDX/AL/WAX: 74/21/5

EXPLOSIVE PROPERTIES

NUMBER FORT EFORM EXPLOSIVE COMPOSITION BY WEIGHT

	KCAL/G	C	H	N	O	AL
17	1.300	.020360	.163	.027	.260	.320 .210

VENTING CALCULATION

CHARGE WEIGHT(LB) = .2940E-01
INIT VOLUME(CU FT) = 0.000
INIT VENT AREA(SQ FT) = .5450E-02
AMBIENT PRESSURE(PSIA) = 14.70
AMBIENT TEMP(C) = 20.00
CHAMBER PRESSURE(PSIA) = 14.70
CHAMBER TEMP(C) = 20.00
NOPT= 1 NV= 2

BEGIN VENTING CALCULATION

TABLE OF VOLUME AND VENT AREA CHANGES

P(PSIA)	T(SEC)	V(CU FT)	A(SQ FT)	PAWB(PSIA)	TAMB(C)	NOPTV
0.	.1500	4.000	.5450E-02	14.70	20.00	2
0.	.6000	4.000	.5450E-02	14.70	20.00	2

PROPERTIES OF GASES--

OXIDATION COMPLETE
TEMPERATURE, DEGREES F = 1653.2
ENERGY RELEASE(KCAL/G) = 3.6573
SPECIFIC HEAT RATIO = 1.3141
GAS OVERPRESSURE(PSI) = 45.945

BEGIN VENTING OF GASES

OVERPR(PSI)	TIME(SEC)	GASES(LB)	TEMP(R)	GAMMA	MEON
45.94	0.	.0167	2113.	1.3141	
41.35	.6936E-01	.5809	2074.	1.3155	1
36.76	.1453	.5443	2032.	1.3167	1
36.49	.1500	.5422	2029.	1.3169	1

TIME HAS REACHED TV(1)= .1500

FAILURE LEVEL IN TABLE EXCEEDED.

VOLUME INCREASE(CU FT)	NEW TOT VOL (CU FT)	NEW TOT AREA (SQ FT)	NEW PRESSURE(PSIA)	NEW GAMMA
4.000	12.00	.1090E-01	36.75	1.343
22.05	.1500	.0417	1400.	1.3430
19.85	.1904	.0039	1370.	1.3464
17.64	.2504	.7654	1355.	1.3470
15.44	.3066	.7264	1330.	1.3492
13.23	.3674	.6864	1304.	1.3508
12.69	.3831	.6740	1290.	1.3513
10.49	.4512	.6361	1270.	1.3530
8.203	.5273	.5945	1240.	1.3550
6.420	.6000	.5587	1212.	1.3570

TIME HAS REACHED TV(2)= .6000

FAILURE LEVEL IN TABLE EXCEEDED.

VOLUME INCREASE(CU FT)	NEW TOT VOL (CU FT)	NEW TOT AREA (SQ FT)	NEW PRESSURE(PSIA)	NEW GAMMA
4.000	16.00	.1635E-01	17.06	1.386
2.350	.6000	.0582	848.7	1.3064
2.122	.6170	.0496	845.4	1.3080
1.886	.6351	.0410	842.1	1.3083
1.651	.6543	.0324	838.7	1.3085
1.415	.6750	.0237	835.3	1.3088
1.179	.6976	.0151	831.9	1.3091
.9431	.7226	.0063	828.4	1.3094
.7076	.7511	.0076	824.9	1.3097
.4716	.7850	.0088	821.3	1.3099
.2350	.8296	.0099	817.7	1.3093
.2350E-01	.9084	.0719	814.4	1.3096

FIG. 7.5 OUTPUT RESULTS FOR EXAMPLE 6

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INTERNAL BLAST DAMAGE MECHANICS PROGRAM, MAR 1972
RDX/AL/WAX: 74/21/5

EXPLOSIVE PROPERTIES

NUMBER FOUR EFFNM EXPLOSIVE COMPOSITION BY WEIGHT
KCAL/O C H N O AL
17 1.300 .029760 .163 .027 .240 .120 .210

VENTING CALCULATION

CHARGE WEIGHT(LB) = .2940E-01
INIT VOLUME(CU FT) = 8.000
INIT VENT AREA(SQ FT) = .5450E-02
AMBIENT PRESSURE(Psia) = 14.70
AMBIENT TEMP(C) = 20.00
CHAMBER PRESSURE(Psia) = 14.70
CHAMBER TEMP(C) = 20.00
NOPT= 1 NV= 3

BEGIN VENTING CALCULATION

TABLE OF VOLUME AND VENT AREA CHANGES

P(Psia)	T(SEC)	V(CU FT)	A(SQ FT)	PAMB(Psia)	TAMB(C)	NOPTV
45.00	.1500	4.000	.5450E-02	14.70	20.00	3
70.00	.6000	4.000	.5450E-02	14.70	20.00	3
19.00	.4000	4.000	0.	14.70	20.00	3

PROPERTIES OF GASES--

OXIDATION COMPLETE
TEMPERATURE, DEGREES F = 1453.2
ENERGY RELEASE(KCAL/O) = 3.6573
SPECIFIC HEAT RATIO = 1.3141
GAS OVERPRESSURE(Psia) = 45.945

BEGIN VENTING OF GASES

OVERPR(Psia)	TIME(SEC)	GASES(LB)	TEMP(R)	GAMMA NEON
45.94	0.	.6167	2113.	1.3141
41.35	.6936E-01	.5809	2074.	1.3155 1
36.76	.1453	.5443	2032.	1.3167 1
36.40	.1500	.5422	2029.	1.3169 1

TIME HAS REACHED TV(1)= .1500

FAILURE LEVEL IN TABLE EXCEEDED.

VOLUME INCREASE(CU FT)= 4.000
NEW TOT VOL (CU FT) = 12.00
NEW TOT AREA (SQ FT) = .1090E-01
NEW PRESSURE(Psia) = 36.75
NEW GAMMA = 1.343

22.05	.1500	.6417	1400.	1.3430 1
19.85	.1984	.6039	1370.	1.3464 1
17.64	.2504	.7654	1355.	1.3478 1
15.44	.3066	.7264	1330.	1.3492 1
13.23	.3674	.6866	1304.	1.3506 1
12.49	.3831	.6768	1298.	1.3513 1
10.40	.4512	.6361	1270.	1.3530 2
8.201	.5273	.5945	1240.	1.3550 2
6.424	.6000	.5587	1212.	1.3570 2

TIME HAS REACHED TV(2)= .6000

FAILURE LEVEL IN TABLE EXCEEDED.

VOLUME INCREASE(CU FT)= 4.000
NEW TOT VOL (CU FT) = 16.00
NEW TOT AREA (SQ FT) = .1635E-01
NEW PRESSURE(Psia) = 17.06
NEW GAMMA = 1.386

2.354	.6000	.8582	848.7	1.3864 2
2.122	.6170	.8496	845.4	1.3880 2
1.886	.6351	.8410	842.1	1.3893 2
1.651	.6543	.8324	838.7	1.3905 2
1.415	.6750	.8237	835.3	1.3918 2
1.179	.6976	.8151	831.9	1.3931 2
.9431	.7226	.8063	828.4	1.3944 2
.7074	.7511	.7976	824.9	1.3957 2
.4714	.7850	.7888	821.3	1.3969 2
.2354	.8296	.7799	817.7	1.3981 2
.2354E-01	.9004	.7719	814.4	1.3994 2

FIG. 7.6 OUTPUT RESULTS FOR EXAMPLE 7

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INTERNAL BLAST DAMAGE MECHANISMS PROGRAM, MAR 1972

EXPLOSIVE PROPERTIES

NUMBER EQWT EFORM EXPLOSIVE COMPOSITION BY WEIGHT
 KCAL/G C H N O AL
 17 1.300 .029360 .163 .027 .280 .320 .210
 CHARGE SHAPE CORRECTION IS CRUDE. PSI EXCEEDS
 RANGE OF EXPERIMENTAL DATA.
 CASE WEIGHT CORRECTION IS CRUDE. PSI EXCEEDS
 RANGE OF EXPERIMENTAL DATA.

SHOCK WAVE CALCULATION

INPUT PARAMETERS

CHARGE WEIGHT (LB) = .2940E-01
 EXPLOSIVE NUMBER = 17
 L/D RATIO = 2.700
 CASE/CHARGE WT RATIO = 4.240
 CHAMBER PRESSURE (PSIA) = 14.70
 CHAMBER TEMP (C) = 20.00
 ALTITUDE (KFT) = 0.

CHARGE WEIGHT ADJUSTMENTS

ADJUSTED WT (LB TNT) = .6318E-01
 HE ENERGY FACTOR = 1.300
 CHARGE SHAPE FACTOR = 2.894
 CASE WEIGHT FACTOR = .5711
 PRESSURE SCALE FACTOR = .9997
 DISTANCE SCALE FACTOR = 2.511
 TIME SCALE FACTOR = 2.489
 NORMAL REFL FACTOR = 6.752

DESIRED DISTANCE (FT) = .6670
 (CM) = 20.33

TIME AFTER EXPLOSION (MSEC)	TIME AFTER SHOCK ARR (MSEC)	INCIDENT OVERPRESS (PSI)	NORM REFL OVERPRESS (PSI)
7.1180E-02	0.	317.6	2144.
9.5353E-02	2.4173E-02	100.2	676.2
.1074	3.6259E-02	63.11	426.1
.1195	4.8345E-02	40.98	276.7
.1316	6.0431E-02	26.81	181.0
.1437	7.2518E-02	17.27	116.6
.1558	8.4604E-02	10.64	71.84
.1679	9.6690E-02	5.914	39.93
.1800	.1088	2.494	16.84
.1920	.1209	0.	0.

IMPULSE (PSI.MSEC)--

INCIDENT = 7.675
 REFLECTED = 51.82

CAUTION--CONTACT SURFACE HAS ARRIVED.

DATA ARE CRUDE BEYOND T(MSEC) AFTER SHOCK ARRIVAL = 2.7874E-02

FIG. 7.7 OUTPUT RESULTS FOR EXAMPLE 8

INTERNAL BLAST DAMAGE MECHANISMS PROGRAM, MAR 1972

EXPLOSIVE PROPERTIES

NUMBER LGWT EFORM EXPLOSIVE COMPOSITION BY WEIGHT
 KCAL/G C H N O AL
 17 1.300 .024360 .163 .027 .280 .320 .210
 CHARGE SHAPE CORRECTION IS CRUDE. PSI EXCEEDS
 RANGE OF EXPERIMENTAL DATA.
 CASE WEIGHT CORRECTION IS CRUDE. PSI EXCEEDS
 RANGE OF EXPERIMENTAL DATA.

SHOCK WAVE CALCULATION

INPUT PARAMETERS

CHARGE WEIGHT (LB) = .2940E-01
 EXPLOSIVE NUMBER = 17
 L/D RATIO = 2.700
 CASE/CHARGE WT RATIO = 4.240
 CHAMBER PRESSURE (PSIA) = 6.760
 CHAMBER TEMP (C) = -24.60
 ALTITUDE (KFT) = 0.

CHARGE WEIGHT ADJUSTMENTS

ADJUSTED WT (LB TNT) = .7408E-01
 HE ENERGY FACTOR = 1.300
 CHARGE SHAPE FACTOR = 3.394
 CASE WEIGHT FACTOR = .5711
 PRESSURE SCALE FACTOR = 2.174
 DISTANCE SCALE FACTOR = 1.638
 TIME SCALE FACTOR = 1.707
 NORMAL REFL FACTOR = 7.852

DESIRED DISTANCE (FT) = .6670
 (CM) = 20.33

TIME AFTER EXPLOSION (MSEC)	TIME AFTER SHOCK ARR (MSEC)	INCIDENT OVERPRESS (PSI)	NORM REFL OVERPRESS (PSI)
5.8490E-02	0.	276.5	217.1
8.5063E-02	2.6573E-02	87.14	684.7
9.8349E-02	3.9859E-02	54.94	431.4
.1116	5.3146E-02	35.88	280.1
.1249	6.6432E-02	23.34	183.2
.1382	7.9719E-02	15.04	118.1
.1515	9.3005E-02	9.264	72.74
.1648	.1063	5.148	40.42
.1781	.1196	2.171	17.05
.1914	.1329	0.	0.

IMPULSE (PSI.MSEC)--

INCIDENT = 7.345
 REFLECTED = 57.67

CAUTION --CONTACT SURFACE HAS ARRIVED.

DATA ARE CRUDE BEYOND T(MSEC) AFTER SHOCK ARRIVAL = 1.5305E-02

FIG. 7.8 OUTPUT RESULTS FOR EXAMPLE 9

REFERENCES

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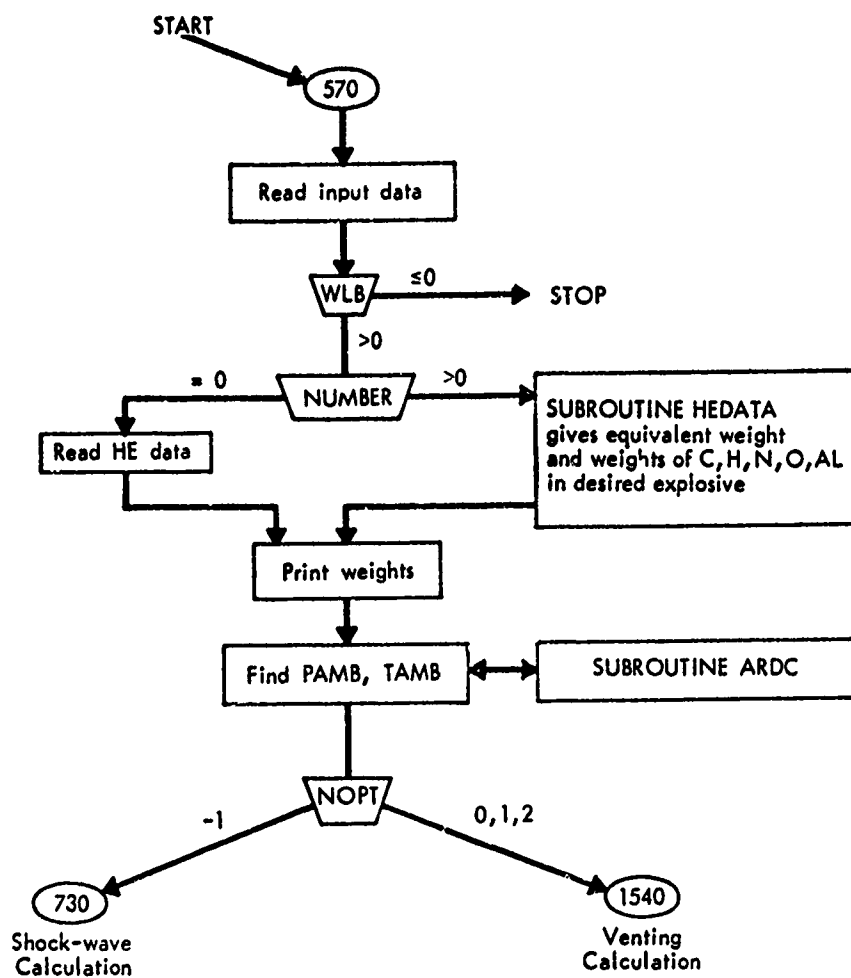
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APPENDIX A

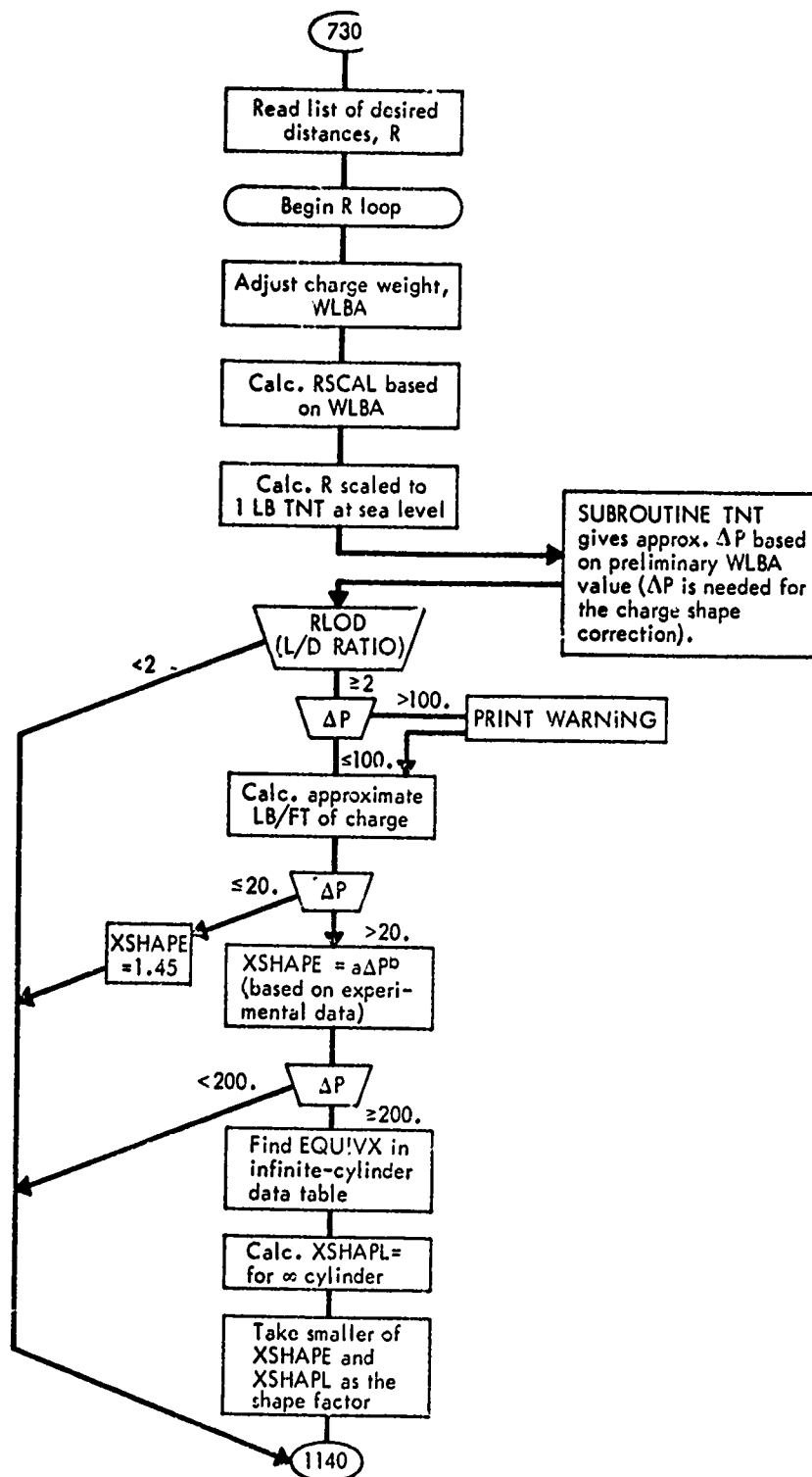
FLOW CHART FOR COMPUTER CODE

The following pages of this appendix give the complete flow chart for the computer program. It is broken into three logical sections, input, shock wave calculations, and venting calculations.

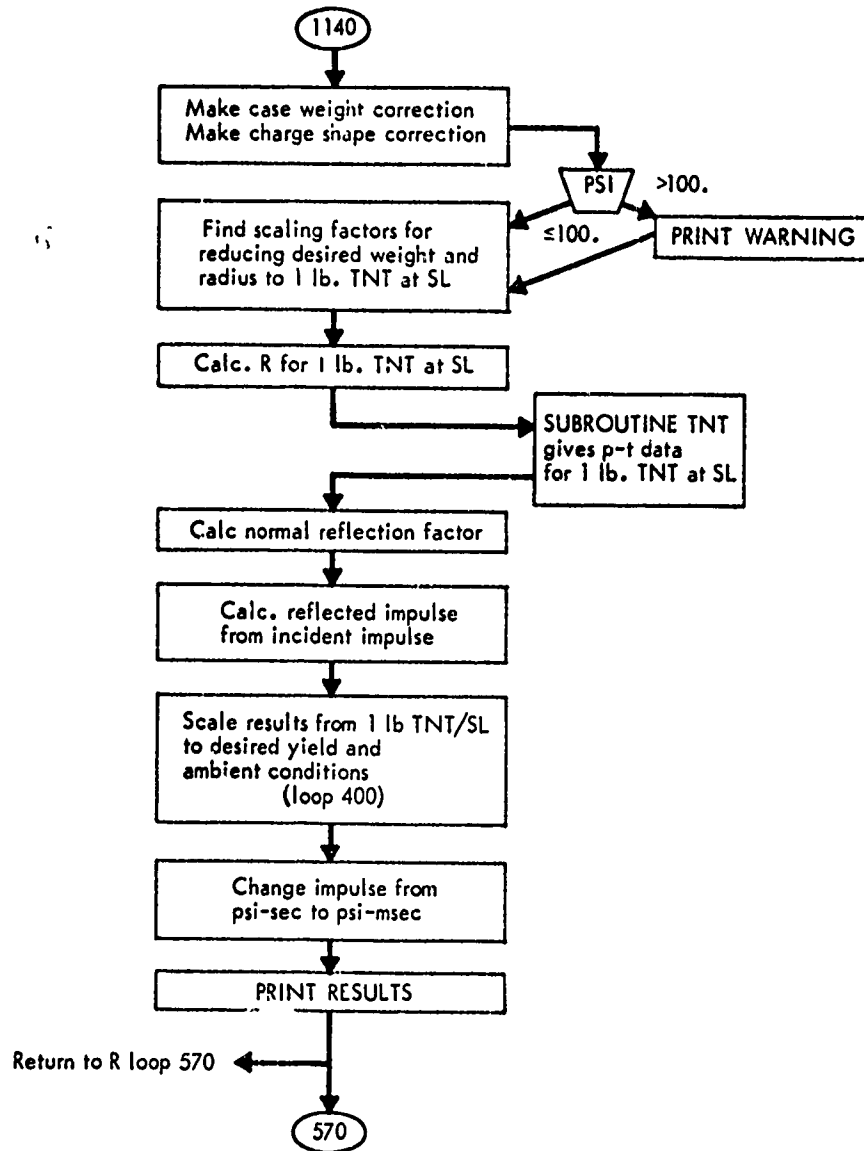
INPUT



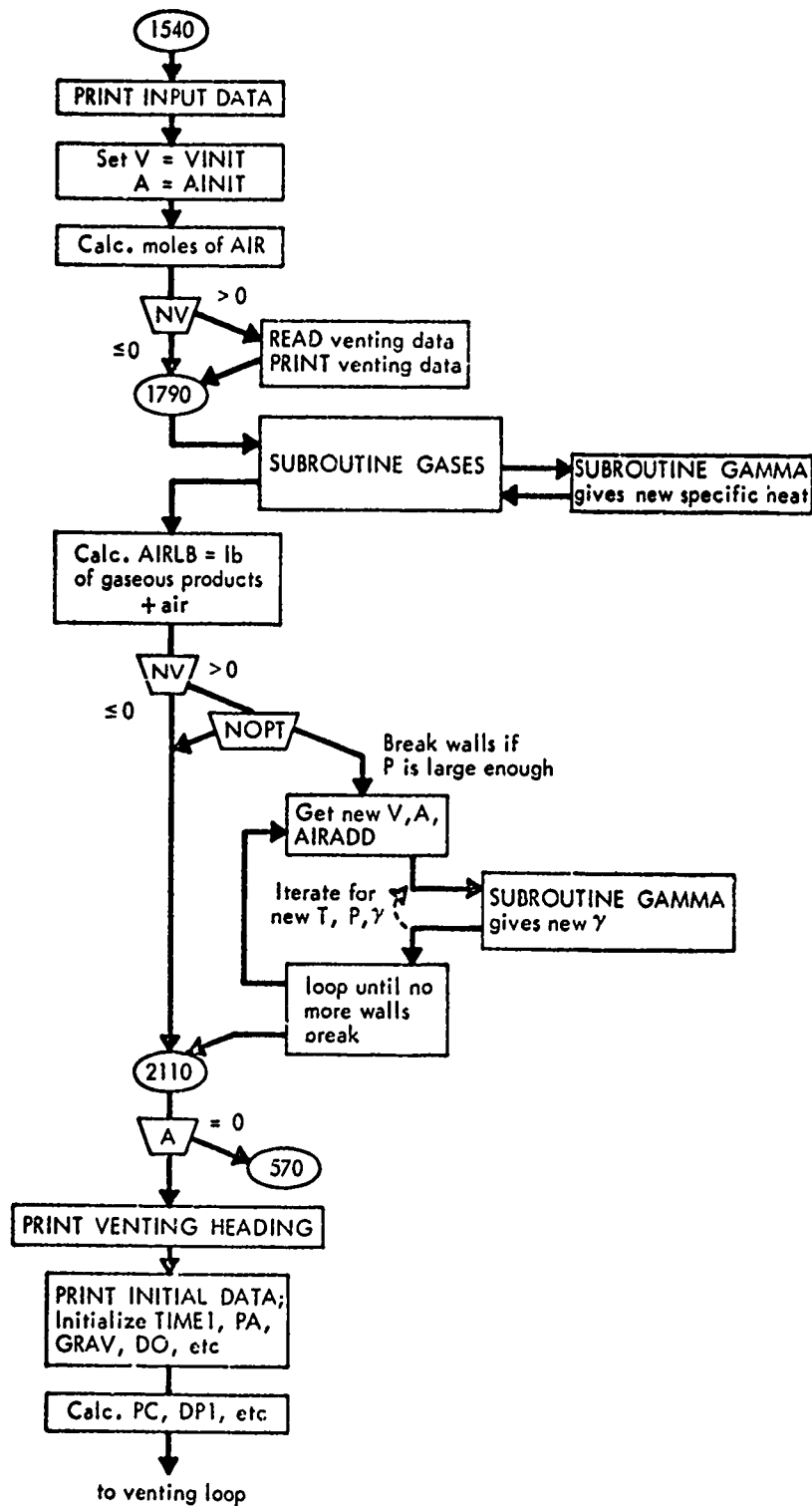
SHOCK-WAVE CALCULATION



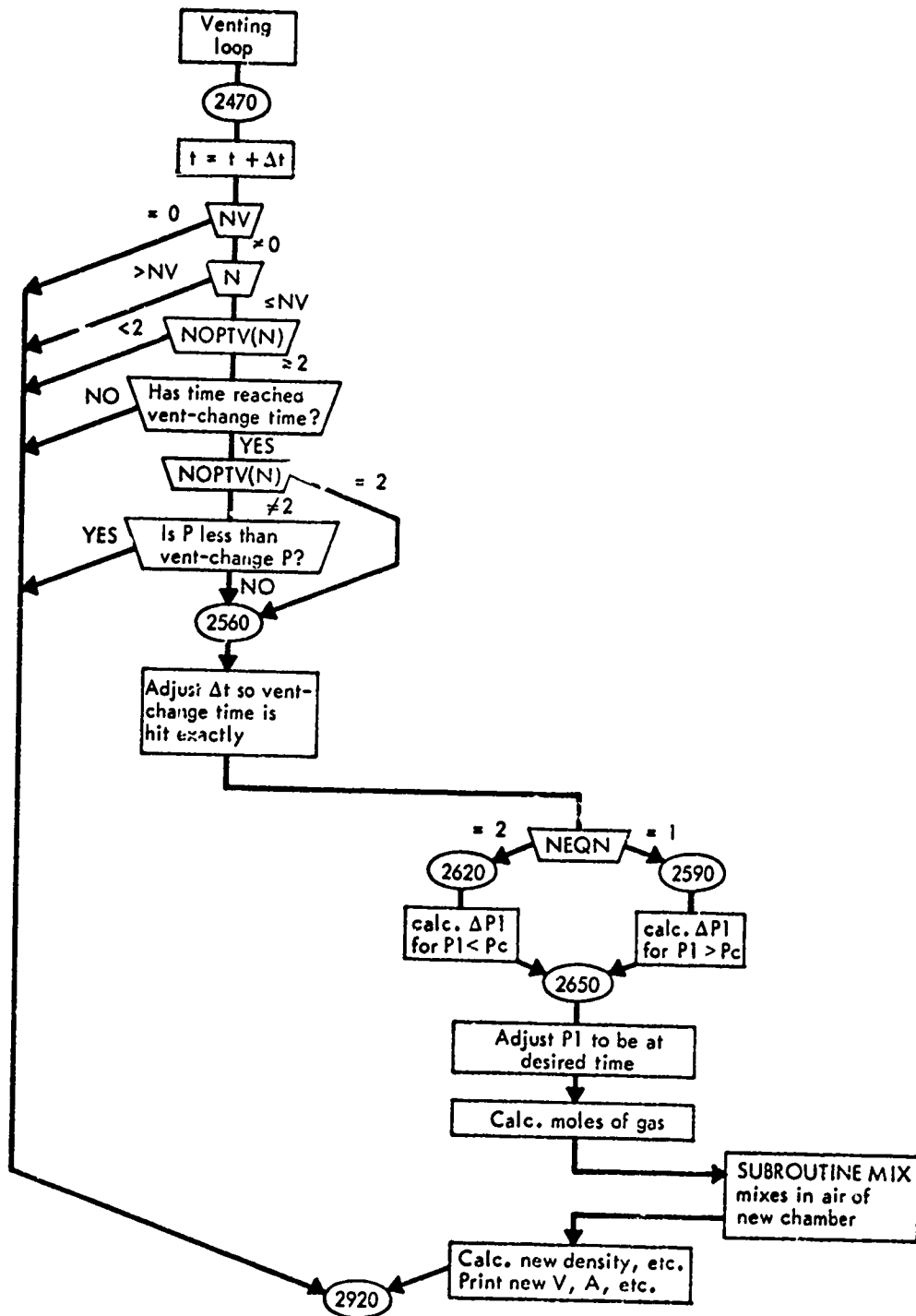
SHOCK-WAVE CALCULATION (CONT'D)



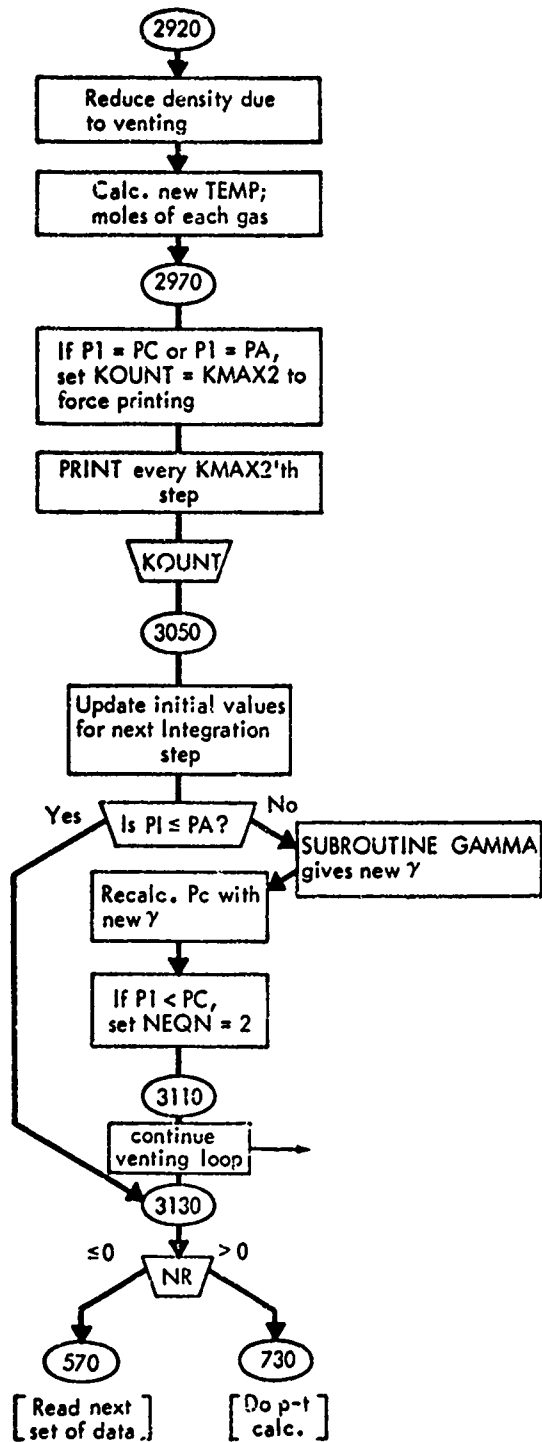
VENTING CALCULATION



VENTING CALCULATION (CONT'D)



VENTING CALCULATION (CONT'D)



APPENDIX B

INPUT DATA CARDS

This appendix provides descriptions and explanations of all of the information required on the input data cards for this program. Formats of these cards are given, and the sample problem input cards in Figure 7.1 can be used as guides.

DESCRIPTION OF INPUT DATA CARDS

FIRST DATA CARD--FORMAT(E5.2, I5, 9E5.2, 3I5)

NOTE THAT THE THREE QUANTITIES INDICATED BY ---*--- MAY CAUSE ADDITIONAL CARDS TO BE READ IN.

WLB =WEIGHT OF EXPLOSIVE CHARGE (POUNDS).

*NUMBER=IDENTIFICATION NUMBER OF DESIRED EXPLOSIVE IN LIST.

IF THE DESIRED EXPLOSIVE IS NOT IN THE LIST, EITHER

(1) USE THE NEAREST AVAILABLE ONE IN THE LIST, OR

(2) ADD THE NEW EXPLOSIVE TO THE LIST, OR

(3) READ IN THE DESIRED PROPERTIES AFTER THIS CARD.

RLOD =LENGTH/DIAMETER RATIO.

CASE =CASE WEIGHT/EXPLOSIVE WEIGHT RATIO.

VINIT =INITIAL VOLUME OF CHAMBER (.CUBIC FEET).

AINIT =INITIAL VENT AREA (SQUARE FEET).

PAMB =AMBIENT PRESSURE INTO WHICH VENTING OCCURS (PSIA).

TAMB =AMBIENT TEMPERATURE INTO WHICH VENTING OCCURS (CENTIGRADE).

ALTKFT=ALTITUDE (KILOFEET).

IF PAMB AND TAMB ARE BOTH GIVEN AS 0., THE CORRECT VALUES WILL BE FOUND BY THE PROGRAM FROM THE ARDC ATMOSPHERE.

ALTKFT IS IGNORED IF PAMB AND TAMB ARE NOT 0.

PCHAM =INITIAL AMBIENT PRESSURE IN CHAMBER (PSIA).

TCHAM =INITIAL AMBIENT TEMPERATURE IN CHAMBER (C).

IF PCHAM AND TCHAM ARE 0., THEY ARE ASSUMED TO EQUAL PAMB AND TAMB.

NOPT =1 DO VENTING CALC. =2 DO SHOCK P-T CALC.

*NV =NUMBER OF CARDS OF VENTING CHANGE DATA TO BE READ IN.

*NR =NUMBER OF RADII AT WHICH SHOCK P-T DATA ARE WANTED.

SECOND DATA CARD. OMIT THIS CARD IF NUMBER IS POSITIVE.

THIS CARD HAS TWO POSSIBLE FORMS DEPENDING ON WHETHER NUMBER IS 0 OR -1.

IF NUMBER=0, READ IN THE FOLLOWING EXPLOSIVE DATA FOR AN EXPLOSIVE NOT APPEARING IN THE LIST IN SUBROUTINE HEDATA. FORMAT(7E7.2)

WFACT =BLAST EQUIVALENCE RELATIVE TO TNT (USUALLY ABOUT 1.0).

EFORM =ENERGY OF FORMATION OF THE EXPLOSIVE (CAL/GRAM).

WFC =WEIGHT FRACTION CARBON.

WFH =WEIGHT FRACTION HYDROGEN.

WFN =WEIGHT FRACTION NITROGEN.

WFO =WEIGHT FRACTION OXYGEN.

WFA =WEIGHT FRACTION ALUMINUM.

(NOTE THAT THESE ARE WEIGHT FRACTION, NOT WEIGHT PERCENT.)

IF NUMBER=-1, READ IN THE FOLLOWING DATA FOR PREPARING A MIXTURE OF THE COMPONENTS IN THE LIST IN SUBROUTINE HEDATA. FORMAT(E7.2,9(I3,F4.3)).

WFACT =BLAST EQUIVALENCE RELATIVE TO TNT (USUALLY ABOUT 1.0).

NUMHE (1)=EXPLOSIVE NUMBER IN THE TABLES.

HEFRAC(1)=WEIGHT FRACTION OF THIS EXPLOSIVE.

NUMHE (2)=SAME FOR SECOND COMPONENT.

HEFRAC(2)=SAME FOR SECOND COMPONENT.

... CONTINUE FOR AS MANY AS 9 COMPONENTS.

THIRD DATA CARD(S). OMIT IF NV=0. FORMAT(6E7.2,17)
 THERE IS ONE CARD PER N, THERE ARE NV OF THESE CARDS.
 THIS IS AN ARRAY OF VENT AREA AND VOLUME CHANGES.
 PV(N) =PRESSURE AT WHICH A-V CHANGE IS TO OCCUR (PSIA). IF THE INITIAL
 CHAMBER PRESSURE EXCEEDS PV(N), A NEW CHAMBER IS ADDED.
 TV(N) =TIME AT WHICH A-V CHANGE IS TO OCCUR (SEC).
 VV(N) =NEW VOLUME TO BE ADDED (CUBIC FEET).
 AV(N) =NEW VENT AREA TO BE ADDED (SQUARE FEET).
 PAV(N)=AMBIENT PRESSURE IN NEW VOLUME (PSIA).
 TAV(N)=AMBIENT TEMPERATURE IN NEW VOLUME (C).
 NOPTV(N)=CONTROLS USE OF VENT AREA AND VOLUME CHANGE TABLES.
 =1 BREAK INTO NEW VOLUME IF INITIAL PRESSURE EXCEEDS PV(N).
 =2 BREAK INTO NEW VOLUME IF TIME TV(N) IS REACHED.
 =3 BREAK INTO NEW VOLUME IF PRESSURE EXCEEDS PV(N) WHEN TIME
 TV(N) IS REACHED.

FOURTH DATA CARD(S). OMIT IF NR=0. FORMAT (10E7.2)
 THERE ARE NR/10 OF THESE CARDS WITH 10 R VALUES PER CARD.
 TOTAL OF NR ELEMENTS IN ARRAY.
 R(I) =ARRAY OF DESIRED RADII AT WHICH SHOCK P-T DATA IS WANTED (FT).

APPENDIX C

DEFINITIONS OF PROGRAM VARIABLES

A complete alphabetical listing of all program variables used in this code is given in this appendix. Also a definition accompanies each listed variable.

DEFINITIONS OF PROGRAM VARIABLES

A =CURRENT VALUE OF VENT AREA(SQ FT).
 AINIT =INITIAL VENT AREA(SQ FT).
 AIRADD=LB MOLES OF AIR IN NEWLY ADDED CHAMBER.
 AIRMOL=LB MOLES OF AIR IN ALL ACTIVE CHAMBERS.
 ALTKFT=ALTITUDE(KILOFEET). NOT USED IF PAMB, TAMB ARE GIVEN.
 AV(N) =ARRAY OF NEW VENT AREAS(SQ FT).
 ALTH =GEOPOTENTIAL ALTITUDE(METERS).
 ALTZ =ALTITUDE(METERS) ABOVE MEAN SEA LEVEL.
 CASE =CASE WEIGHT/CHARGE WEIGHT RATIO.
 DIF1 =PREVIOUS DIFFERENCE BETWEEN 2 WAYS TO CALC P.
 DIF2 =CURRENT DIFFERENCE BETWEEN 2 WAYS TO CALC P.
 DPI =INTEGRATION INTERVAL IN PRESSURE(PSE).
 DO =ENERGY ADDED(KCAL) IN INTEGRATION STEP.
 DT =INTEGRATION INTERVAL IN FINDING TEMPERATURE.
 DTEMP =TEMPERATURE(R) INTERVAL IN T,P,G ITERATION.
 DTIME1=TIME INTERVAL IN VENTING INTEGRATION (SEC).
 DUM =DUMMY VARIABLE IN SUBROUTINE CALL.
 DO =DENSITY(LB/CU FT) AT START OF VENTING STEP.
 DI =DENSITY(LB/CU FT) AT END OF VENTING STEP.
 EF =ARRAY OF HE ENERGY OF FORMATION DATA (KCAL/G).
 EFORM =ENERGY OF FORMATION OF DESIRED EXPLOSIVE (KCAL/G).
 EQUIVX=INTERMEDIATE QUANTITY IN CHARGE SHAPE CORRECTION.
 EQUIV2=ARRAY OF CYL-SPH EQUIVALENCE FACTORS.
 EQWT =EQUIVALENT WEIGHT REFERRED TO TNT (FOR SHOCK CALCS).
 FA =ARRAY OF WEIGHT FRACTION AL.
 FC =ARRAY OF WEIGHT FRACTION C.
 FH =ARRAY OF WEIGHT FRACTION H.
 FN =ARRAY OF WEIGHT FRACTION N.
 FO =ARRAY OF WEIGHT FRACTION O.
 FLEFT =LOSS FRACTION FOR VENTING MASS CHANGES.
 FRAC =INTERPOLATION FACTOR.
 G =CURRENT VALUE OF SPECIFIC HEAT RATIO.
 GASLB =POUNDS OF GAS REMAINING IN ACTIVE CHAMBERS.
 GASMOL=LB MOLES OF GAS REMAINING IN ACTIVE CHAMBERS.
 GG =(G-1.)/G
 GRAV =ACCELERATION OF GRAVITY =32.2 FT/SEC/SEC.
 G0 =SPECIFIC HEAT RATIO AT START OF INTEGRATION STEP.
 G1 =SPECIFIC HEAT RATIO AFTER MIXING GASES.
 G2 =SPECIFIC HEAT RATIO.
 HEFRAC=ARRAY OF WEIGHT FRACTIONS (USED WITH NUMHE).
 HF =ACCELERATION FOR HEFRAC(I).
 I =GENERAL DO-LOOP INDEX.
 IR =INDEX FOR DO-LOOP ON R(IR). RANGE IS 1 TO NR.
 J =DO-LOOP INDEX
 JJ =DESIRED TNT DATA LIE BETWEEN R(JJ) AND R(JJ-1).
 K =DO-LOOP INDEX

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KMAX1 = DESIRED NUMBER OF P-T POINTS BEFORE HE GASES ARRIVE.
 TYPICAL VALUES ARE 10 TO 40.
 KMAX2 = VENTING PRINTOUT CONTROL. PRINT ABOUT 100/KMAX2 LINES OF
 DATA. TYPICAL VALUES ARE 2 TO 10.
 KOUNT = COUNTER FOR PRINTING DURING VENTING.
 L = DO-LOOP INDEX
 MCO = LB MOLES OF CO IN THE CHAMBER.
 MCO2 = LB MOLES OF CO2 IN THE CHAMBER.
 MH2 = LB MOLES OF H2 IN THE CHAMBER.
 MH2O = LB MOLES OF H2O IN THE CHAMBER.
 MN2 = LB MOLES OF N2 IN THE CHAMBER.
 MO2 = LB MOLES OF O2 IN THE CHAMBER.
 M1 = GRAM MOLES OF CO2 FORMED.
 M2 = GRAM MOLES OF H2 FORMED.
 M3 = NOT USED.
 M4 = NOT USED.
 M5 = GRAM MOLES OF AL2O3 FORMED.
 M6 = NOT USED.
 M7 = NOT USED.
 M8 = GRAM MOLES OF H2O FORMED.
 M9 = GRAM MOLES OF CO FORMED.
 NAME = ARRAY FOR NAMES OF EXPLOSIVES.
 NAMES = ARRAY FOR NAME OF DESIRED EXPLOSIVE.
 N = EXPLOSIVE NUMBER IN TABLE, OR CHAMBER BREAKING INDEX.
 NN = INDEX FOR VENTING LOOP.
 NEQN = 1 FOR P.GT.PC, = 2 FOR P.LT.PC (CHOOSSES VENTING EQUATION).
 NOPT = 1 DO VENTING CALC, = 2 DO SHOCK P-T CALC.
 NOPTV(N) = ARRAY OF WALL-BREAKING OPTIONS.
 = 1 BREAK WALL IF PRESSURE EXCEEDS PV(N).
 = 2 BREAK WALL WHEN TIME TV(N) IS REACHED.
 = 3 BREAK WALL IF PRESSURE EXCEEDS PV(N) AT TIME TV(N).
 NR = NUMBER OF ELEMENTS IN USE IN R ARRAY (1 TO 100).
 NSAVE = LAST LINE OF VENT DATA USED IN INITIAL BREAKS.
 NUMBER = NUMBER OF DESIRED EXPLOSIVE IN DATA LIST.
 NUMHE = ARRAY OF EXPLOSIVE NUMBERS FOR ARBITRARY MIXING UP OF HE.
 NV = NUMBER OF ELEMENTS IN VENTING ARRAY.
 N1 = GRAM MOLES OF C IN THE EXPLOSIVE.
 N2 = GRAM MOLES OF H2 IN THE EXPLOSIVE.
 N3 = GRAM MOLES OF N2 IN THE EXPLOSIVE.
 N4 = GRAM MOLES OF O2 IN THE EXPLOSIVE.
 N5 = GRAM MOLES OF AL IN THE EXPLOSIVE.
 N6 = GRAM MOLES OF N2 IN THE CHAMBER AIR.
 N7 = GRAM MOLES OF O2 IN THE CHAMBER AIR.
 OVERP2 = ARRAY OF CYL-SPH EQUIVALENCE OVERPRESSURES (PSI).
 OVPSI = OVERPRESSURE (PSI) AT START OF VENTING.
 OVPSI1 = CURRENT OVERPRESSURE (PSI).
 OVPO = OVERPRESSURE (PSI) AT START OF VENTING STEP.
 OVPI = OVERPRESSURE (PSI) AT END OF VENTING STEP.

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P(I) =ARRAY OF INCIDENT OVERPRESSURE(PSI) FOR TNT.
 PA =INITIAL OVERPRESSURE(PSI) IN CHAMBER WITHOUT GAMMA CORR.
 PAMB =OUTSIDE AMBIENT PRESSURE(PSIA).
 PAV(N)=ARRAY OF AMBIENT PRESSURES(PSIA) IN NEW CHAMBERS.
 PC(JJ)=ARRAY OF CONTACT SURFACE ARRIVAL TIME(SEC) DATA FOR TNT.
 PCHAM =INITIAL AIR PRESSURE(PAIA) IN ORIGINAL CHAMBER.
 PCTNT =CONTACT SURFACE OVERPRESSURE(PSI) AT RTNT.
 PI =OVERPR/PEAK OVERP RATIO FROM FITTING EQUATION.
 PINIT =INITIAL CHAMBER PRESSURE(PSFA) BEFORE WALLS BREAK.
 PSCAL =SCALING FACTOR FOR REDUCING PRESSURES TO SEA LEVEL.
 PSFAMB=OUTSIDE AMBIENT PRESSURE(PSFA).
 PSFCH =CHAMBER PRESSURE(PSFA).
 PSI(I)=ARRAY OF OVERPRESSURE(PSI) AT DESIRED RADIUS R AT TIME T1(I).
 PSIREF(I)=REFLECTED OVERPRESSURE(PSI) CORRESP. TO PSI(I).
 PTNT =PEAK OVERPRESSURE(PSI) AT RTNT.
 PTSCL=PSCAL*TSCL
 PV(N) =ARRAY OF PRESSURES(PSIA) WALLS CAN WITHSTAND.
 PO =PRESSURE(PSFA) AT START OF VENTING STEP.
 POPSI =PO IN OVERPRESSURE(PSI).
 P1 =PRESSURE(PSFA) AT END OF VENTING STEP.
 P1PSI =PRESSURE(PSIA) AT END OF VENTING STEP.
 P2 =PRESSURE(PSFA) AFTER WALL BREAKS.
 P2A =PRESSURE FROM FIRST EQN (FOR MIXING TWO CHAMBERS).
 P2B =PRESSURE FROM SECOND EQN (FOR MIXING TWO CHAMBERS).
 Q =ENERGY(KCAL) RELEASED BY EXPLOSION.
 QPERG =ENERGY RELEASED(KCAL/GRAM).
 Q1 =CUMULATIVE ENERGY DURING INTEGRATION FOR TEMPERATURE.
 R =GRAM MOLES O2 LEFT IN CHAMBER (CALLED RR IN BLAST).
 R(I) =ARRAY OF DESIRED DISTANCES(FT). NR ELEMENTS IN THIS ARRAY.
 RCM =DISTANCE R(I) CONVERTED TO CM.
 REF =OVERPRESSURE REFLECTION FACTOR.
 RESULT =QUANTITIES BEING PRINTED IN SUBROUTINE GASES.
 RLOD =LENGTH/DIAMETER RATIO OF CHARGE.
 RR =(IN BLAST ONLY) GRAM MOLES OF O2 LEFT IN THE CHAMBER.
 RSCAL =SCALING FACTOR FOR REDUCING RADII TO SEA LEVEL.
 RTNT =R(I)P REDUCED TO 1 LB TNT AT SEA LEVEL.
 SIGMA =SHAPE PARAMETER IN FITTING EQUATION FOR P-T DATA FOR TNT.
 T =TEMPERATURE(R) IN SUBROUTINE GAMMA.
 TAMB =OUTSIDE AMBIENT TEMPERATURE(C).
 TAU =FRACTION OF POSITIVE DURATION BASED ON 65. CM CURVE SHAPE.
 TAV(N)=ARRAY OF AMBIENT TEMPS(C) IN NEW CHAMBERS.
 TCHAM =INITIAL TEMP(C) IN ORIGINAL CHAMBER BEFORE EXPLOSION.
 TC(JJ)=ARRAY OF TIME(SEC) BETWEEN SHOCK AND CS ARRIVAL FOR TNT.
 TCMSEC=TCNT IN MSEC.
 TCTNT =CONTACT SURFACE ARRIVAL TIME(SEC) AT RTNT.
 TEMP =TEMPERATURE IN ARDC SUBROUTINE.
 TEMPO =GAS TEMP(R) AT START OF VENTING STEP.
 TEMP1 =GAS TEMP(R) AT END OF VENTING STEP.
 TEMP2 =GAS TEMP(R) AFTER NEW CHAMBER IS ADDED.
 TF =GAS TEMPERATURE(FAHRENHEIT).

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TIME1 = TIME(SEC) AT END OF VENTING STOP.
 TOTMOL = TOTAL POUND MOLES OF GASES IN CHAMBER.
 TP(JJ) = ARRAY OF POSITIVE OVERPR DURATION(SEC) FOR TNT.
 TPTNT = POSITIVE PHASE DURATION(SEC) AT RTNT.
 TR = GAS TEMP(R) IN CHAMBER.
 TRCH = AMBIENT TEMP(R) IN ORIGINAL CHAMBER BEFORE EXPLOSION.
 TS(JJ) = ARRAY OF SHOCK FRONT ARRIVAL TIME(SEC) DATA FOR TNT.
 TSCAL = SCALING FACTOR FOR REDUCING TIMES TO SEA LEVEL.
 TSTAT = SHOCK FRONT ARRIVAL TIME(SEC) AT RTNT.
 TV(N) = ARRAY OF TIMES(SEC) FOR BREAKING WALLS.
 T1(I) = ARRAY OF TIME(SEC) AFTER DETONATION.
 T2(I) = ARRAY OF TIME(SEC) AFTER SHOCK ARRIVAL.
 U = SPECIFIC HEAT OF GAS MIXTURE.
 U1 = SPECIFIC HEAT OF CO2.
 U2 = SPECIFIC HEAT OF H2.
 U4 = SPECIFIC HEAT OF O2.
 U6 = SPECIFIC HEAT OF N2.
 U8 = SPECIFIC HEAT OF H2O.
 U9 = SPECIFIC HEAT OF CO.
 V = ACTIVE VOLUME(CU FT).
 VINIT = INITIAL CHAMBER VOLUME(CU FT).
 VV(N) = ARRAY OF NEW CHAMBER VOLUMES(CU FT).
 V0 = CHAMBER VOLUME(CU FT) AT START OF VENTING STEP.
 V1 = CHAMBER VOLUME(CU FT) AT END OF VENTING STEP.
 V2 = CHAMBER VOLUME(CU FT) AFTER NEW CHAMBER IS ADDED.
 WA = POUNDS OF AL IN THE EXPLOSIVE.
 WC = POUNDS OF C IN THE EXPLOSIVE.
 WFA = WEIGHT FRACTION OF AL IN THE EXPLOSIVE.
 WFACT = CHARGE ENERGY RELATIVE TO EQUAL WEIGHT OF TNT.
 WFC = WEIGHT FRACTION OF C IN THE EXPLOSIVE.
 WFH = WEIGHT FRACTION OF H IN THE EXPLOSIVE.
 WFN = WEIGHT FRACTION OF N IN THE EXPLOSIVE.
 WFO = WEIGHT FRACTION OF O IN THE EXPLOSIVE.
 WFT = APPROXIMATE CHARGE LENGTH(FT).
 WH = POUNDS OF H IN THE EXPLOSIVE.
 WLB = WEIGHT(LB) OF EXPLOSIVE CHARGE.
 WLBA = ADJUSTED CHARGE WEIGHT(LB).
 WN = POUNDS OF N IN THE EXPLOSIVE.
 WO = POUNDS OF O IN THE EXPLOSIVE.
 WPERL = APPROXIMATE CHARGE WEIGHT PER UNIT LENGTH(LB/FT).
 XCASE = CORRECTION FACTOR FOR CASE EFFECT.
 XIMPI = SIDE-ON POSITIVE IMPULSE BEFORE HE GAS ARRIVAL (PSI.MSEC).
 XIMPIR = REFLECTED OVERPR POSITIVE IMPULSE BEFORE GAS ARR (PSI.MSEC).
 XSHAPE = CORRECTION FACTOR FOR CHARGE CYLINDRICITY.
 XSHAPL = CORRECTION FACTOR FOR INFINITE LINE CHARGE.
 X2 = GRAM MOLES OF GAS IN CHAMBER.
 X3 = GRAM MOLES OF GAS IN CHAMBER WITHOUT THE AIR.

APPENDIX D

FORTRAN LISTING OF PROGRAM

This appendix gives the complete FORTRAN listing of the computer program. All seven sections are labeled with appropriate card numbers.

BLAS0010 - BLAS3160
MIX 0010 - MIX 0270
HEDA0010 - HEDA1400
GAMM0010 - GAMM0160
GAS 0010 - GAS 0890
TNT 0010 - TNT 1330
ARDC0010 - ARDC0410

D-2

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1      * RANGE OF EXPERIMENTAL DATA.*
KNAX1=10  KNAX2=10
C
C READ INPUT DATA.
570 READ 300, IL, NUMBER, RLCD, CASE, VINIT, AINIT, PAMB, TAMB, ALTAFT,
1 PCHAM, TCHAM, NOPT, NV, NR
IF (RLCD.EQ.0) STOP  SPRINT -20  SIF (NUMBER.EQ.0) GO TO 600
IF (NUMBER.EQ.-1) GO TO 594
READ 305, IFACT, (NUMP(I), HLEFAC(I), I=1,9)
596 CALL MEDATA  SGO TO 605
600 READ 310, EFACT, EFORM, EFC, EFH, EFN, EFO, EFA  SEFORM=EFORM/1000.
C POUNDS OF EACH ELEMENT
605 WC=EFC*ALB  SAA=EFH*ALB  SAA=EFH*ALB  SAU=EFU*ALB  SAA=EFA*ALB
IF (EFACT.GT.0.) GO TO 615
EFACT=1.  SPRINT 612
612 FORMAT(* EFACT NOT KNOWN, 1.0 IS USED.*)
615 PRINT 620, NUMBER, EFACT, EFORM, EFC, EFH, EFN, EFO, EFA
620 FORMAT(*EXPLOSIVE PROPERTIES*/
1* NUMBER EXP. EFORM EXPLOSIVE COMPOSITION BY WEIGHT*/
2*          KCAL/G      C      H      N      O      AL */
31H ,I4,F7.3,F8.5,5F6.3)
C FIND PAMB AND TAMB IF NOT GIVEN.
IF (PAMB.EQ.0.) CALL ARSC(ALTAFT, PAMB, TAMB)
IF (PCHAM.LE.0.) PCHAM=PAMB  SIF (TCHAM.LE.0.) TCHAM=TAMB
C DO VENTING CALC IF NOPT=1 AND DO SHOCK WAVE CALC IF NOPT=2.
IF (NOPT.LT.2) GO TO 1540
C
C BEGIN SHOCK-WAVE PROPERTIES SECTION.
730 READ 310, (IR, I=1, NR)
DO 1480 IR=1, NR  SIF (IR.EQ.1) GO TO 760
PRINT 420  SPRINT 620, NUMBER, EFACT, EFORM, EFC, EFH, EFN, EFO, EFA
760 RCM=R(IR)*30.48
C ADJUST CHARGE WEIGHT.
WLBA=WL3*EFACT  SPAMB=PCHAM  STAMB=TCHAM
RSCAL=(1./WLBA*PAMB/14.696175)**.33333333
RTNT=RCM*RSCAL  SCALL THT(C)
C MAKE CHARGE SHAPE CORRECTION.
XSHAPE=1.  SIF (RLCD.LT.2.) GO TO 1140
IF (PSI(1).GT.100.) PRINT 535
C CHARGE WEIGHT PER UNIT LENGTH OF CYLINDRICAL CHARGE.
WPERL=(3.1416*100./(4.*RLCD**2))**.33333333 *WLBA**.66666666
C MAKE CHARGE SHAPE CORRECTION.
IF (PSI(1).GT.20.) GO TO 900  XSHAPE=1.45  SGO TO 1140
900 XSHAPE=0.513*PSI(1)**.287
C FIND INFINITE-CYLINDER CHARGE SHAPE CORRECTION.
IF (PSI(1).LT.200.) GO TO 1140
DO 940 I=2, 24  SIF (PSI(1).LT.OVERP2(I)) GO TO 950
940 CONTINUE  I=24
950 FRAC=(PSI(1)-OVERP2(I-1))/(OVERP2(I)-OVERP2(I-1))
EQUIVX=EQUIV2(I-1)+FRAC*(EQUIV2(I)-EQUIV2(I-1))
XSHAPL=EQUIVX*WPERL**1.5/WLBA
IF (XSHAPL.LT.XSHAPE) XSHAPL=XSHAPL
C TEST IF DESIRED DISTANCE IS CLOSE ENOUGH TO CHARGE FOR GOOD RESULTS.
C APPROX. CHARGE LENGTH.

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BLAS0542
 BLAS0560
 BLAS0562
 BLAS0565
 BLAS0570
 BLAS0572
 BLAS0574
 BLAS0576
 BLAS0580
 BLAS0596
 BLAS0600
 BLAS0602
 BLAS0605
 BLAS0608
 BLAS0610
 BLAS0612
 BLAS0615
 BLAS0620
 BLAS0625
 BLAS0630
 BLAS0640
 BLAS0650
 BLAS0660
 BLAS0670
 BLAS0690
 BLAS0700
 BLAS0710
 BLAS0720
 BLAS0730
 BLAS0740
 BLAS0750
 BLAS0760
 BLAS0770
 BLAS0780
 BLAS0800
 BLAS0810
 BLAS0830
 BLAS0840
 BLAS0850
 BLAS0860
 BLAS0870
 BLAS0880
 BLAS0890
 BLAS0900
 BLAS0910
 BLAS0920
 BLAS0930
 BLAS0940
 BLAS0950
 BLAS0960
 BLAS0970
 BLAS0980
 BLAS1010
 BLAS1020

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1030 *FT=KPERL*VLB
C MAKE CASE WEIGHT CORRECTION. CASE=CASE/CHARGE WEIGHT RATIO.
1140 XCASE=0.47+0.53/(1.+CASE)
      IF(CASE.LT.0.53) XCASE=1.-CASE**2/(1.+CASE)
      WLBPA=WLBH*XSHPAE*XCASE
      IF(PST(1).GT.100.) PRINT 540
C FIND SCALING FACTORS.
      PSCAL=14.696178/PAVB
      RSCAL=(1./WLBPA*PAMB/14.696178)**.33333333
      TSCAL=RSCAL*SQRT((273.16+TAMB)/288.16)
C FIND DESIRED RADIUS FOR 1 LB TNT AT SEA LEVEL.
      RTNT=RC/RSCAL
C FIND DATA IN TABLES.
      CALL TNT(1)
C CALCULATE NORMAL REFLECTION FACTOR.
      IF(PST(1).GT.200.) GO TO 1280
      REF=(7.*14.696178+4.*PST(1))/(7.*14.696178+PST(1))*2.5GO TO 1290
1280 REF=-3.18+3.97*ALOG10(PST(1)) 3IF(REF.GT.13.) REF=13.
1290 XIMP1=XIMP1*REF
C SCALE RESULTS TO DESIRED CHARGE WEIGHT AND AMBIENT CONDITIONS.
C CHANGE TIMES FROM SECONDS TO MSEC.
      DO 1350 K=1,KMAX1
      PSI(K)=PSI(K)/PSCAL  $PSIREF(K)=PSI(K)*REF
      T1(K)=T1(K)/TSCAL*1000.  $T2(K)=T2(K)/TSCAL*1000.
1350 CONTINUE
      PTSCAL=PSCAL*TSCAL
      XIMP1=XIMP1*1000./PTSCAL  $XIMP1R=XIMP1R*1000./PTSCAL
C PRINT THE RESULTS.
      PRINT 320,WLB,WLBPA,NUMSER,*FACT,RL0D,XSHPAE,CASE,XCASE,
      1 PAMB,PSCAL,TAMB,RSCAL,ALTAFT,TSCAL,REF
      PRINT 430,R(IR),RCM  $PRINT 450
      PRINT 490,(T1(K),T2(K),PSI(K),PSIREF(K),K=1,KMAX1)
      PRINT 500,XIMP1,XIMP1R  $TCMSEC=TCNT*1000./TSCAL
      IF(RTNT.LT.65.) PRINT 520,TCMSEC
1480 CONTINUE
      GO TO 570
C END SHOCK-WAVE PROPERTIES SECTION.
C
C
C BEGIN VENTING SECTION.
1540 PRINT 1550,WLB,VINIT,AINIT,PAVB,TAMB,PCHAM,TCHAM,NCPT,NV
1550 FORMAT(*OVENTING CALCULATION*/
      1*CHARGE WEIGHT(LB)  **G12.4/
      3* INIT VOLUME(CU FT)  **G12.4/
      4* INIT VENT AREA(SQ FT)  **G12.4/
      5* AMBIENT PRESSURE(PSIA)=*G12.4/
      6* AMBIENT TEMP(C)  **G12.4/
      7* CHARGE PRESSURE(PSIA)=*G12.4/
      8* CHARGE TEMP(C)  **G12.4/
      9* NCPT=*I3,* NV=*I3)
1650 FORMAT(6E7.0,I7)
1660 FORAT(*TABLE OF VOLUME AND VENT AREA CHANGES*/
      1* P(PSIA)  T(SEC)  V(CU FT)  A(SQ FT)  PAMB(PSIA)  TAMB(PSIA)
      2*3(C) NCPTV*/10(IH,6G12.4,I7))
1680 FORAT(*BEGIN VENTING CALCULATION*)

```

ELAS1030
 ELAS1130
 ELAS1140
 ELAS1150
 ELAS1160
 ELAS1162
 ELAS1170
 ELAS1180
 ELAS1190
 ELAS1200
 ELAS1210
 ELAS1220
 ELAS1230
 ELAS1240
 ELAS1250
 ELAS1260
 ELAS1270
 ELAS1280
 ELAS1290
 ELAS1300
 ELAS1310
 ELAS1320
 ELAS1330
 ELAS1340
 ELAS1350
 ELAS1355
 ELAS1360
 ELAS1380
 ELAS1390
 ELAS1400
 ELAS1410
 ELAS1420
 ELAS1430
 ELAS1440
 ELAS1480
 ELAS1490
 ELAS1500
 ELAS1510
 ELAS1520
 ELAS1530
 ELAS1540
 ELAS1550
 ELAS1560
 ELAS1580
 ELAS1590
 ELAS1600
 ELAS1610
 ELAS1620
 ELAS1630
 ELAS1640
 ELAS1650
 ELAS1660
 ELAS1670
 ELAS1680
 ELAS1690

```

      A=AINIT
      PRINT 1690
      PSFCH=PCHAN*144.   STRCH=(TCNAM+273.16)*1.8   SPSFAMB=PAMB*144.
C POUND MOLES OF AIR.
      AIRPOL=PSFCH*VINIT/(1545.*TRCH)
      IF(NV.LE.0) GO TO 1790
      READ 1650,(PV(N),TV(N),VV(N),AV(N),PAV(N),TAV(N),NOPTV(N),/I=1,NV)
      PRINT1660,(PV(N),TV(N),VV(N),AV(N),PAV(N),TAV(N),NOPTV(N),N=1,NV)
1790 CALL GASES(VINIT,AIR*CL,OVPC,GC,TR)
      GASLB=32.*MO2+ 28.*MN2+ 28.*CO+ 44.*MCO2+ 18.*MH2O+ 2.*MH2
      GASMOL=MO2+MN2+MCO+MCO2+MH2O+MH2
C PINIT=INITIAL CHAMBER PRESSURE (PSFA) AFTER EXPLOSION.
      PINIT=OVPC*144. +PSFCH
      PO=PINIT   SVO=VINIT   STEMP0=TR   SNSAVE=1
      IF(NV.LE.0) GO TO 2110
      IF(NOPTV(1),NE.1) GO TO 2110
C
C BREAK WALLS IF PO EXCEEDS TABULATED VALUES.
      DO 2070 N=1,NV   SNSAVE=N
      IF(NOPTV(N),NE.1) GO TO 2110
      IF(PO.LT.PV(N)*144.) GO TO 2070
      A=AV(N)
      CALL MIX(PO,V0,TEMP0,G0,GASMOL, P2,V2,TEMP2,G2,AIRADD)
      PO=P2   SVO=V2   STEMP0=TEMP2   SG0=G2
C PO, V0, TEMP0, G0 ARE NOW AFTER NEW VOLUME IS ADDED.
C ADD NEW AIR TO MOLES OF N2 AND O2.
      MN2=MN2+.7901*AIRADD   SMO2=MO2+.2095*AIRADD
      GASMOL=MO2+MN2+MCO+MCO2+MH2O+MH2
      POPSI=PO/144.
      PRINT 2010,VV(N),V2,A,POPSI,G2
2010 FORMAT(10FAILURE LEVEL IN TABLE EXCEEDED.*/
1* VOLUME INCREASE(CU FT)=*G12.4/
2* NEW TOT VOL (CU FT)   =*G12.4/
3* NEW TOT AREA (SQ FT)  =*G12.4/
4* NEW PRESSURE(PSIA)    =*G12.4/
5* NEW GAMMA              =*G12.4)
2070 CONTINUE
C INITIAL BREAKING INTO NEW CHAMBERS IS NOW COMPLETED.
C
C NO VENTING IF AREA=0.
2110 IF(A.EQ.0.) GO TO 570
      PO=INITIAL PEAK PRESSURE (PSFA).
      V0=INITIAL VOLUME (CU FT).
      TO=INITIAL TEMP(R).
      PRINT 2160
2160 FORMAT(10BEGIN VENTING OF GASES*/
1* OVERPR(PSI)   TIME(SEC)   GASES(LB)   TEMP(R)   GAMMA NEQN*)
      OVPSI=(PO-PSFAMB)/144.   STIME1=0.   SG=G0
      GASLB=32.*MO2+ 28.*MN2+ 28.*MCO+ 44.*MCO2+ 18.*MH2O+ 2.*MH2
      PRINT 3030,OVPSI,TIME1,GASLB,TEMP0,G
      PA=PSFAMB   SGRAV=32.2
C DENSITY (LB/CU FT).
      DO=GASLB/V0
C CRITICAL PRESSURE (PSFA).

```

BLAS1700
 BLAS1710
 BLAS1720
 BLAS1730
 BLAS1740
 BLAS1750
 BLAS1760
 BLAS1770
 BLAS1780
 BLAS1790
 BLAS1800
 BLAS1810
 BLAS1820
 BLAS1830
 BLAS1840
 BLAS1850
 BLAS1860
 BLAS1870
 BLAS1880
 BLAS1890
 BLAS1900
 BLAS1910
 BLAS1920
 BLAS1930
 BLAS1940
 BLAS1950
 BLAS1960
 BLAS1970
 BLAS1980
 BLAS1990
 BLAS2000
 BLAS2010
 BLAS2020
 BLAS2030
 BLAS2040
 BLAS2050
 BLAS2060
 BLAS2070
 BLAS2080
 BLAS2090
 BLAS2100
 BLAS2110
 BLAS2120
 BLAS2130
 BLAS2140
 BLAS2150
 BLAS2160
 BLAS2170
 BLAS2180
 BLAS2190
 BLAS2200
 BLAS2210
 BLAS2220
 BLAS2230
 BLAS2240

```

PC=PA/((2./(G+1.))*((G/(G-1.)))
C PRESSURE INCREMENT
DP1=(P0-PA)*.01
NEON=1 $TIME1=0. SKOUNT=1
V1=V0
C
N=NSAVE
C BEGIN VENTING LOOP.
DO 3110 NN=1,1000 SP1=P0-DP1 SG=GO
IF(P1.GT.PA) GO TO 2360
KOUNT=99 SP1=P0 SGO TO 3010
2360 GO TO (2380,2430),NEON
C VENTING FOR P1.GT.PC
2380 IF(P1.LE.PC.AND.P0.GT.PC) P1=PC
DTIME1=(P0-P1)/P1**((3.*G-1.)/(2.*G))*V1/A/
1 SQRT(GRAV*G**3*P0**((1./G)/DO**((2./(G+1.))*((G+1.)/(G-1.)))
GO TO 2470
C VENTING FOR P1.LT.PC
2430 IF(P1.LT.PA.AND.P0.GT.PA) P1=PA
GG=(G-1.)/G
DTIME1=(PC-P1)/(P1**GG*(P1**GG-PA**GG)**.5)*V1/A/
1 SQRT(GRAV*G**3*2./(G-1.)*(PC*PA**2/DO**G)**((1./G))
2470 TIME1=TIME1+DTIME1
IF(NV.EQ.0) GO TO 2920
IF(N.GT.NV) GO TO 2920
IF(NOPTV(N).LT.2) GO TO 2920
C CHECK TIME AGAINST VENTING TABLE.
IF(TIME1.LT.TV(N)) GO TO 2920
IF(NOPTV(N).EQ.2) GO TO 2560
IF(P1.LT.PV(N)*144.) GO TO 2920
C ADJUST TIME1 TO EQUAL TV.
2560 DTIME1=TV(N)-(TIME1-DTIME1)
TIME1=TV(N)
GO TO (2590,2620),NEON
2590 DP1 =P1**((3.*G-1.)/(2.*G))*A/V1*DTIME1*
1 SQRT(GRAV*G**3*(P0**((1./G)/GO**((2./(G+1.))*((G+1.)/(G-1.)))
GO TO 2650
2620 GG=(G-1.)/G
DP1=P1**GG*(P1**GG-PA**GG)**.5*A/V1*DTIME1*
1 SQRT(GRAV*G**3*(2./(G-1.)*(P0*PA**2/DO**G)**((1./G))
2650 P1=P0-DP1 SCVP1=P1-PA IOVPS11=CVP1/144.
C REDUCE MASSES DUE TO VENTING.
D1=DO*(P1/PC)**((1./G)
TEMP1=TEMP0*(P1/P0)**((G-1.)/G)
C FRACTION LEFT AFTER THIS VENTING STEP.
FLEFT=D1/DO
MO2=MO2+FLEFT $MH2=$MH2+FLEFT $MCO=$MCO+FLEFT
MCO2=$MCO2+FLEFT $MH20=$MH20+FLEFT $MH2=$MH2+FLEFT
GASLB=D1*V1
GASMO=$O2+M12+MCO+MCO2+MH20+MH2
PRINT 3030,OVPS11,TIME1,GASLB,TEMP1,G,NEON
PRINT 2770,N,TIME1
2770 FORMAT(* TIME HAS REACHED TV(*12*)=*G12.4)
C BEGIN VOLUME-AREA CHANGE SECTION.
A=A+AV(N)

```

BLAS2250
 BLAS2260
 BLAS2270
 BLAS2280
 BLAS2290
 BLAS2300
 BLAS2310
 BLAS2320
 BLAS2330
 BLAS2340
 BLAS2350
 BLAS2360
 BLAS2370
 BLAS2380
 BLAS2390
 BLAS2400
 BLAS2410
 BLAS2420
 BLAS2430
 BLAS2440
 BLAS2450
 BLAS2460
 BLAS2470
 BLAS2480
 BLAS2490
 BLAS2500
 BLAS2510
 BLAS2520
 BLAS2530
 BLAS2540
 BLAS2550
 BLAS2560
 BLAS2570
 BLAS2580
 BLAS2590
 BLAS2600
 BLAS2610
 BLAS2620
 BLAS2630
 BLAS2640
 BLAS2650
 BLAS2660
 BLAS2670
 BLAS2680
 BLAS2690
 BLAS2700
 BLAS2710
 BLAS2720
 BLAS2730
 BLAS2740
 BLAS2750
 BLAS2760
 BLAS2770
 BLAS2780
 BLAS2790

CALL MIX(P1,V1,TEMP1,G,GASMOL, P2,V2,TEMP2,G2,AIRADD)	BLAS2800
P1=P2 \$V1=V2 \$TEMP1=TEMP2 \$G=G1=G2	BLAS2810
MH2=MN2+.7005*AIRADD	BLAS2820
MO2=MC2+.2095*AIRADD	BLAS2830
GASLB=32.*MO2+ 28.*MN2+ 28.*MCO+ 44.*MCO2+ 18.*MH2O+ 2.*MH2	BLAS2840
V1=V2 \$DPSI=P1/144.	BLAS2850
D1=GASLB/V1 \$DP1=(P1-PA)*.01	BLAS2860
PRINT 2010,VV(N),V1,A,P1PSI,G1	BLAS2870
N=N+1 \$KOUNT=KMAX2 \$GO TO 2970	BLAS2880
C END VOLUME-AREA CHANGE SECTION.	BLAS2890
C	BLAS2900
C REDUCE MASSES DUE TO VENTING.	BLAS2910
2920 D1=D0*(P1/P0)**(1./G)	BLAS2920
TEMP1=TEMP0*(P1/P0)**((G-1.)/G)	BLAS2930
FLEFT=D1/D0 \$GASLB=D1*V1	BLAS2940
MO2=MO2*FLEFT \$MN2=MN2*FLEFT \$MCO=MCO*FLEFT	BLAS2950
MC2=MC2*FLEFT \$MH2O=MH2O*FLEFT \$MH2=MH2*FLEFT	BLAS2960
2970 IF(P1.EQ.PC) KOUNT=KMAX2	BLAS2970
IF(P1.EQ.PA) KOUNT=KMAX2	BLAS2980
C PRINT EVERY KMAX2-TH LINE.	BLAS2990
IF(KOUNT.NE.KMAX2) GO TO 3050 \$KOUNT=0	BLAS3000
3010 OVPSI1=(P1-PA)/144.	BLAS3010
PRINT 3030,OVPSI1,TIME1,GASLB,TEMP1,G,NEON	BLAS3020
3030 FORMAT(1H ,4G12.4,F7.4,14)	BLAS3030
IF(KOUNT.EQ.99) GO TO 3130	BLAS3040
3050 P0=P1 \$KOUNT=KOUNT+1 \$D0=D1 \$TEMP0=TEMP1	BLAS3050
IF(P1.LE.PA) GO TO 3130	BLAS3060
CALL GAMMA(TEMP1,GO,DUM) \$G=GO	BLAS3070
C RECALCULATE PC WITH NEW G.	BLAS3080
IF(P1.GT.PC) PC=PA/((2./(G+1.))**(G/(G-1.)))	BLAS3090
IF(P1.LE.PC) NEON=2	BLAS3100
5110 CONTINUE	BLAS3110
C END VENTING LOOP.	BLAS3120
3130 IF(NR.GT.0) GO TO 730	BLAS3130
GO TO 570	BLAS3140
C END VENTING SECTION.	BLAS3150
END	BLAS3160

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SUBROUTINE MIX(P0,V0,TEMP0,G0,GASMOL, P2,V2,TEMP2,G2,AIRADD)      MIX 0010
C MIX THE GASES IN ADJACENT CHAMBERS.                             MIX 0020
COMMON/VENT/ PV(10),TV(10),VV(10),AV(10),PAV(10),TAV(10),NOPTV(10)MIX 0030
1,N                                                                MIX 0040
V2=V0+VV(N)                                                       MIX 0050
C MOLES OF AIR IN THE NEW VOLUME.                                  MIX 0060
AIRADD=(PAV(N)*144.)*VV(N)/(1545.*(TAV(N)+273.16)*1.8)          MIX 0070
C ITERATE TO FIND NEW T,P,G.                                       MIX 0080
TOTMOL=GASMOL+AIRADD                                             MIX 0090
DIF1=1.E10  $DTEMP=TEMP0/100.                                     MIX 0100
C P0, V0, TEMP0, G0 ARE BEFORE NEW VOLUME IS ADDED.              MIX 0110
C P2, V2, TEMP2, G2 ARE AFTER NEW VOLUME IS ADDED.               MIX 0120
C FIRST GUESS FOR TEMP2.                                          MIX 0130
TEMP2=TEMP0                                                       MIX 0140
DO 240 J=1,100                                                    MIX 0150
P2A=(TOTMOL*1545./V2)*TEMP2                                       MIX 0160
CALL GAMMA(TEMP2,G2,DUM)                                          MIX 0170
P2B=(G2-1.)/V2 *(((1.4-1.)*P0*V0 +(G0-1.)*PAV(N)*VV(N))/      MIX 0180
1 ((1.4-1.)*(G0-1.))                                              MIX 0190
DIF2=ABS(P2A-P2B)                                                 MIX 0200
IF(DIF2.GT.DIF1) GO TO 250                                        MIX 0210
C CONTINUE SEARCH FOR CORRECT TEMP2.                               MIX 0220
TEMP2=TEMP2-DTEMP  $DIF1=DIF2                                     MIX 0230
240 CONTINUE                                                       MIX 0240
250 P2=(P2A+P2B)/2.                                               MIX 0250
RETURN                                                            MIX 0260
END                                                                MIX 0270

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[illegible]

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DATA((NAME(12,1),I=1,5)=10H(37H TNETB,10H/AL, 72/28,10H	•HEDA0315
110H .10H .13,F6.3))	HEDA0316
DATA EQWT(12),EF(12),FC(12),FH(12),FN(12),FO(12),FA(12)	HEDA0320
1 /1.18, -221.1, .134, .011, .157, .418, .280/	HEDA0330
DATA((NAME(13,1),I=1,5)=10H(37H TNETB,10H/AL, 65/35,10H	•HEDA0335
110H .10H .13,F6.3))	HEDA0336
DATA EQWT(13),EF(13),FC(13),FH(13),FN(13),FO(13),FA(13)	HEDA0340
1 /1.23, -199.6, .121, .010, .142, .377, .350/	HEDA0350
DATA((NAME(14,1),I=1,5)=10H(37H TRITO,10H/AL (TNT/A,10HL,80/20)	•HEDA0355
110H .10H .13,F5.3))	HEDA0356
DATA EQWT(14),EF(14),FC(14),FH(14),FN(14),FO(14),FA(14)	HEDA0360
1 /1.07, -62.72, .296, .018, .148, .338, .200/	HEDA0370
DATA((NAME(15,1),I=1,5)=10H(37H RDX/A,10HL/WAX, 88/10H10/2	•HEDA0375
110H .10H .13,F6.3))	HEDA0376
DATA EQWT(15),EF(15),FC(15),FH(15),FN(15),FO(15),FA(15)	HEDA0380
1 /1.30, 50.38, .160, .027, .333, .380, .100/	HEDA0390
DATA((NAME(16,1),I=1,5)=10H(37H RDX/A,10HL/WAX, 78/10H20/2	•HEDA0395
110H .10H .13,F6.3))	HEDA0396
DATA EQWT(16),EF(16),FC(16),FH(16),FN(16),FO(16),FA(16)	HEDA0400
1 /1.32, 43.76, .144, .024, .295, .337, .200/	HEDA0410
DATA((NAME(17,1),I=1,5)=10H(37H RDX/A,10HL/WAX, 74/10H21/5	•HEDA0415
110H .10H .13,F6.3))	HEDA0416
DATA EQWT(17),EF(17),FC(17),FH(17),FN(17),FO(17),FA(17)	HEDA0420
1 /1.30, 29.36, .163, .027, .280, .320, .210/	HEDA0430
DATA((NAME(18,1),I=1,5)=10H(37H RDX/A,10HL/WAX, 74/10H22/4	•HEDA0435
110H .10H .13,F6.3))	HEDA0436
DATA EQWT(18),EF(18),FC(18),FH(18),FN(18),FO(18),FA(18)	HEDA0440
1 /1.30, 33.28, .154, .026, .280, .320, .220/	HEDA0450
DATA((NAME(19,1),I=1,5)=10H(37H RDX/A,10HL/WAX, 62/10H33/5	•HEDA0455
110H .10H .13,F6.3))	HEDA0456
DATA EQWT(19),EF(19),FC(19),FH(19),FN(19),FO(19),FA(19)	HEDA0460
1 /1.19, 21.42, .143, .024, .235, .268, .330/	HEDA0470
DATA((NAME(20,1),I=1,5)=10H(37H TORPE,10HX 11 (RDX/10HTNT/AL,42/	•HEDA0475
110H40/18) .10H .13,F6.3))	HEDA0476
DATA EQWT(20),EF(20),FC(20),FH(20),FN(20),FO(20),FA(20)	HEDA0480
1 /1.24, -3.57, .216, .021, .233, .350, .180/	HEDA0490
DATA((NAME(21,1),I=1,5)=10H(37H H-6 (10HRDX/TNT/AL,10H/WAX,45/29,	•HEDA0495
110H21/5) .10H .13,F6.3))	HEDA0496
DATA EQWT(21),EF(21),FC(21),FH(21),FN(21),FO(21),FA(21)	HEDA0500
1 /1.27, -12.56, .223, .025, .224, .318, .210/	HEDA0510
DATA((NAME(22,1),I=1,5)=10H(37H HBX-1,10H (RDX/TNT/10HAL/WAX,40/	•HEDA0515
110H38/17/5) .10H .13,F6.3))	HEDA0516
DATA EQWT(22),EF(22),FC(22),FH(22),FN(22),FO(22),FA(22)	HEDA0520
1 /1.21, -22.93, .249, .026, .221, .334, .170/	HEDA0530
DATA((NAME(23,1),I=1,5)=10H(37H HBX-3,10H (RDX/TNT/10HAL/WAX,31/	•HEDA0535
110H29/35/5) .10H .13,F6.3))	HEDA0536
DATA EQWT(23),EF(23),FC(23),FH(23),FN(23),FO(23),FA(23)	HEDA0540
1 /1.16, -21.83, .200, .022, .171, .257, .350/	HEDA0550
DATA((NAME(24,1),I=1,5)=10H(37H TNETB,10H/RDX/AL, 3,10H9/26/35	•HEDA0555
110H .10H .13,F6.3))	HEDA0556
DATA EQWT(24),EF(24),FC(24),FH(24),FN(24),FO(24),FA(24)	HEDA0560
1 /1.24, -102.6, .115, .013, .184, .338, .350/	HEDA0570
DATA((NAME(25,1),I=1,5)=10H(37H ALUMI,10HNUM .10H	•HEDA0575
110H .10H .13,F6.3))	HEDA0576

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DATA EQWT(25),EF(25),FC(25),FH(25),FN(25),FO(25),FA(25)	HEDA0580
1 / 0., 0., 0., 0., 0., 0., 1./	HEDA0590
DATA((NAME(26,I),I=1,5)=10H(37H WAX ,10H ,10H	HEDA0595
110H ,10H ,13,F6.3))	HEDA0596
DATA EQWT(26),EF(26),FC(26),FH(26),FN(26),FO(26),FA(26)	HEDA0600
1 / 0., -392., .856, .144, 0., 0., 0./	HEDA0610
DATA((NAME(27,I),I=1,5)=10H(37H RDX ,10H ,10H	HEDA0615
110H ,10H ,13,F6.3))	HEDA0616
DATA EQWT(27),EF(27),FC(27),FH(27),FN(27),FO(27),FA(27)	HEDA0620
1 / 0., 66.16, .162, .027, .379, .432, 0./	HEDA0630
DATA((NAME(28,I),I=1,5)=10H(37H PETN ,10H ,10H	HEDA0635
110H ,10H ,13,F6.3))	HEDA0636
DATA EQWT(28),EF(28),FC(28),FH(28),FN(28),FO(28),FA(28)	HEDA0640
1 / 0., -407.1, .190, .026, .177, .607, 0./	HEDA0650
DATA((NAME(29,I),I=1,5)=10H(37H TETRY,10HL ,10H	HEDA0655
110H ,10H ,13,F6.3))	HEDA0656
DATA EQWT(29),EF(29),FC(29),FH(29),FN(29),FO(29),FA(29)	HEDA0660
1 / 0., 16.26, .293, .017, .244, .446, 0./	HEDA0670
DATA((NAME(30,I),I=1,5)=10H ,10H ,10H	HEDA0675
110H ,10H ,13,F6.3))	HEDA0676
DATA EQWT(30),EF(30),FC(30),FH(30),FN(30),FO(30),FA(30)	HEDA0680
1 / 0., 0., 0., 0., 0., 0., 0./	HEDA0690
DATA((NAME(31,I),I=1,5)=10H ,10H ,10H	HEDA0695
110H ,10H ,13,F6.3))	HEDA0696
DATA EQWT(31),EF(31),FC(31),FH(31),FN(31),FO(31),FA(31)	HEDA0700
1 / 0., 0., 0., 0., 0., 0., 0./	HEDA0710
DATA((NAME(32,I),I=1,5)=10H ,10H ,10H	HEDA0715
110H ,10H ,13,F6.3))	HEDA0716
DATA EQWT(32),EF(32),FC(32),FH(32),FN(32),FO(32),FA(32)	HEDA0720
1 / 0., 0., 0., 0., 0., 0., 0./	HEDA0730
DATA((NAME(33,I),I=1,5)=10H ,10H ,10H	HEDA0735
110H ,10H ,13,F6.3))	HEDA0736
DATA EQWT(33),EF(33),FC(33),FH(33),FN(33),FO(33),FA(33)	HEDA0740
1 / 0., 0., 0., 0., 0., 0., 0./	HEDA0750
DATA((NAME(34,I),I=1,5)=10H ,10H ,10H	HEDA0755
110H ,10H ,13,F6.3))	HEDA0756
DATA EQWT(34),EF(34),FC(34),FH(34),FN(34),FO(34),FA(34)	HEDA0760
1 / 0., 0., 0., 0., 0., 0., 0./	HEDA0770
DATA((NAME(35,I),I=1,5)=10H ,10H ,10H	HEDA0775
110H ,10H ,13,F6.3))	HEDA0776
DATA EQWT(35),EF(35),FC(35),FH(35),FN(35),FO(35),FA(35)	HEDA0780
1 / 0., 0., 0., 0., 0., 0., 0./	HEDA0790
DATA((NAME(36,I),I=1,5)=10H ,10H ,10H	HEDA0795
110H ,10H ,13,F6.3))	HEDA0796
DATA EQWT(36),EF(36),FC(36),FH(36),FN(36),FO(36),FA(36)	HEDA0800
1 / 0., 0., 0., 0., 0., 0., 0./	HEDA0810
DATA((NAME(37,I),I=1,5)=10H ,10H ,10H	HEDA0815
110H ,10H ,13,F6.3))	HEDA0816
DATA EQWT(37),EF(37),FC(37),FH(37),FN(37),FO(37),FA(37)	HEDA0820
1 / 0., 0., 0., 0., 0., 0., 0./	HEDA0830
DATA((NAME(38,I),I=1,5)=10H ,10H ,10H	HEDA0835
110H ,10H ,13,F6.3))	HEDA0836
DATA EQWT(38),EF(38),FC(38),FH(38),FN(38),FO(38),FA(38)	HEDA0840
1 / 0., 0., 0., 0., 0., 0., 0./	HEDA0850

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DATA((NAME(39,I),I=1,5)=10H          ,10H          ,10H          ,HEDA0855
110H          ,10H ,13,F6.3))          ,HEDA0856
DATA EQWT(39),EF(39),FC(39),FH(39),FN(39),FO(39),FA(39)          ,HEDA0860
1 / 0., 0., 0., 0., 0., 0., 0./          ,HEDA0870
DATA((NAME(40,I),I=1,5)=10H          ,10H          ,10H          ,HEDA0875
110H          ,10H ,13,F6.3))          ,HEDA0876
DATA EQWT(40),EF(40),FC(40),FH(40),FN(40),FO(40),FA(40)          ,HEDA0880
1 / 0., 0., 0., 0., 0., 0., 0./          ,HEDA0890
C          ,HEDA0990
IF(NUMBER.EQ.-1) GO TO 1200          ,HEDA0995
N=NUMBER $WFACT=EQWT(N) $EFORM=EF(N)/1000.          ,HEDA1000
C WEIGHT FRACTIONS          ,HEDA1010
WFC=FC(N) $WFH=FH(N) $WFN=FN(N) $WFO=FO(N) $WFA=FA(N)          ,HEDA1020
DO 1030 L=1,5          ,HEDA1025
1030 NAMES(L)=NAME(N,L) $PRINT NAMES          ,HEDA1030
RETURN          ,HEDA1050
C          ,HEDA1190
C MIX UP AN EXPLOSIVE FROM COMPONENTS IN LIST.          ,HEDA1195
1200 EFORM=WFC+WFH+WFN+WFO+WFA=0. $PRINT 1370          ,HEDA1200
DO 1290 I=1,9 $N=NUMHE(I) $HF=HEFRAC(I)          ,HEDA1210
IF(N.EQ.0) GO TO 1300          ,HEDA1220
DO 1225 L=1,5          ,HEDA1222
1225 NAMES(L)=NAME(N,L) $PRINT NAMES,N,HF          ,HEDA1225
EFORM=EFORM+HF*EF(N) $WFC=WFC+HF*FC(N)          ,HEDA1230
WFH=WFH+HF*FH(N) $WFN=WFN+HF*FN(N)          ,HEDA1240
WFO=WFO+HF*FO(N) $WFA=WFA+HF*FA(N)          ,HEDA1250
1290 CONTINUE          ,HEDA1290
1300 EFORM=EFORM/1000. $RETURN          ,HEDA1300
1370 FORMAT(*0MAKE UP SPECIAL HE MIXTURE---*/          ,HEDA1370
1* NAME*,29X,*NUMBER WT FRAC*)          ,HEDA1375
END          ,HEDA1400

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	SUBROUTINE GAMMA(T,G,U)	GAMM0010
	COMMON/MOLGAS/MO2,MN2,MCO,MCO2,MH2O,MH2	GAMM0020
	REAL MO2,MN2,MCO,MCO2,MH2O,MH2	GAMM0030
C MO2	ETC ARE LB MOLES BUT UNITS CANCEL HERE.	GAMM0040
	U2=5.76+20./T**5+.000578*T	GAMM0060
	U4=11.515-172./T**5+1530./T	GAMM0070
	U6=9.47-3470./T+1160000./T**2	GAMM0080
	U8=19.86-597./T**5+7500./T	GAMM0090
	U9=9.46-3290./T+1070000./T**2	GAMM0100
	U1=16.2-6530./T+1410000./T**2	GAMM0110
	U=U2*MH2+U4*MO2+U6*MN2+U8*MH2O+U9*MCO+U1*MCO2	GAMM0120
	U=U/(MH2+MO2+MN2+MH2O+MCO+MCO2)	GAMM0130
	G=U/(U-1.987)	GAMM0140
	RETURN	GAMM0150
	END	GAMM0160

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SUBROUTINE GASES(V,AIRMOL,P2,G,TR)
C INPUT DATA ARE NPRINT,V,AIRMOL,WC,WH,WN,WO,WA.
C OUTPUT DATA ARE P2,G,TR,Q,M02,MN2,MCO,MCO2,MH2O,MH2.
C V=CHAMBER VOLUME (CU FT).
C AIRMOL=LB MOLES OF AIR IN CHAMBER.
C P2=OVERPRESSURE IN CHAMBER DUE TO HE GASES (PSI).
C G=SPECIFIC HEAT RATIO OF HE GAS-AIR MIXTURE IN CHAMBER.
C TR=TEMPERATURE (RANKINE).
C
C   STATIC CHAMBER PRESSURE CALCULATION ACCOUNTING FOR AVAILABLE O2.
C   COMBUSTION PRODUCT SEQUENCE IS H2O, AL2O3, CO, CO2.
COMMON/DATA1/WLB,NUMBER,RLOD,CASE,VINIT,AINIT,PAMB,TAMB,ALTFT.
1 PCHAM,TCHAM,NOPT,NV,NR, WFACT,EFORM
COMMON/MOLGAS/MO2,MN2,MCO,MCO2,MH2O,MH2
COMMON/HEMASS/WC,WH,WN,WO,WA
COMMON/GAS/N1,N2,N3,N4,N5,N6,N7,
1 M1,M2,M3,M4,M5,M6,M7,M8,M9, R,Q,X2
REAL N1,N2,N3,N4,N5,N6,N7
REAL M1,M2,M3,M4,M5,M6,M7,M8,M9
REAL MO2,MN2,MCO,MCO2,MH2O,MH2
PRINT 210
210 FORMAT(*O PROPERTIES OF GASES--*)
Q=R*M2=M5=M7=M8=M9=M1=0.
C GRAM MOLES C,H2,N2,O2,AL2
N1=WC*453.6/12.
N2=WH*453.6/2.
N3=WN*453.6/28.
N4=WO*453.6/32.
N5=WA*453.6/53.963
N6=N3+.7905*AIRMOL*453.6
N7=N4+.2095*AIRMOL*453.6
C R=GMOL O2 LEFT.
R=N7-N2/2. IF(R.GT.0.) GO TO 340
M8=2.*(R+N2/2.) SM2=N2-M8 SR=0. SQ=57.80*M8 SGO TO 450
340 M8=N2 SQ=N2*57.80 SR=R-1.5*N5 IF(R.GT.0.) GO TO 370
M5=(R+1.5*N5)*(2./3.) SQ=Q+M5*400.3 SR=0. SGO TO 430
370 Q=Q+N5*400.3 SR=R-N1 IF(R.GT.0.) GO TO 510
R=R+N1/2. IF(R.GT.0.) GO TO 420
M9=2.*(R+N1/2.) SQ=Q+M9*26.42 SR=0. SGO TO 470.
420 M1=2.*R SM9=N1-2.*R SQ=Q+M1*94.05+M9*26.42 SR=0. SGO TO 490
430 RESULT=M5/N5 SPRINT 440,RESULT SGO TO 540
440 FORMAT(* PERCENT LAST PRODUCT (AL2O3) ** G12.5)
450 RESULT=M2/N2 SPRINT 460,RESULT SGO TO 540
460 FORMAT(* PERCENT LAST PRODUCT (H2O) ** G12.5)
470 RESULT=M9/N1 SPRINT 480,RESULT SGO TO 540
480 FORMAT(* PERCENT LAST PRODUCT (CO) ** G12.5)
490 RESULT=M1/N1 SPRINT 500,RESULT SGO TO 540
500 FORMAT(* PERCENT LAST PRODUCT (CO2) ** G12.5)
510 PRINT 520
520 FORMAT(* OXIDATION COMPLETE*)
M1=N1 SQ=Q+94.05*N1
540 X2=R+M2+N6+M8+M9+M1
X3=X2-AIRMOL*453.6
Q=Q+.592*X3
C Q=ENERGY RELEASED (KCAL).

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GAS 0010
 GAS 0020
 GAS 0030
 GAS 0040
 GAS 0050
 GAS 0060
 GAS 0070
 GAS 0080
 GAS 0090
 GAS 0100
 GAS 0110
 GAS 0120
 GAS 0130
 GAS 0140
 GAS 0150
 GAS 0160
 GAS 0170
 GAS 0180
 GAS 0190
 GAS 0200
 GAS 0210
 GAS 0220
 GAS 0230
 GAS 0240
 GAS 0250
 GAS 0260
 GAS 0270
 GAS 0280
 GAS 0290
 GAS 0300
 GAS 0310
 GAS 0320
 GAS 0330
 GAS 0340
 GAS 0350
 GAS 0370
 GAS 0400
 GAS 0410
 GAS 0420
 GAS 0430
 GAS 0440
 GAS 0450
 GAS 0460
 GAS 0470
 GAS 0480
 GAS 0490
 GAS 0500
 GAS 0510
 GAS 0520
 GAS 0530
 GAS 0540
 GAS 0550
 GAS 0560
 GAS 0570

Q=Q+WLB*453.6*EFORM	GAS 0580
C LB MOLES OF GASES.	GAS 0600
MO2=R/453.6 \$MN2=N6/453.6 \$MCO=M9/453.6 \$MCO2=M1/453.6	GAS 0610
MH2O=M8/453.6 \$MH2=M2/453.6	GAS 0620
DT=100. \$Q1=0. \$U=7. \$T=1.8*(TCHAM+273.16)	GAS 0630
C INTEGRATE UNTIL T IS FOUND SO ENERGY EQUALS Q.	GAS 0640
DO 760 J=1,100 \$T=T+DT	GAS 0650
CALL GAMMA(T,G,U) \$DQ=(U-1.987)*DT*X2*.0005556	GAS 0660
Q1=Q1+DQ \$IF(Q1.GE.Q) GO TO 780	GAS 0750
760 CONTINUE \$PRINT 770	GAS 0760
770 FORMAT(* T HAS REACHED UPPER LIMIT.*)	GAS 0770
C CORRECT T SO Q1 HITS Q EXACTLY.	GAS 0775
780 T=T-(Q1-Q)/DQ*DT \$G=U/(U-1.987)	GAS 0780
C ABSOLUTE PRESSURE (PSIA)	GAS 0785
P2=X2/453.6*1.987*778./144. *T/V \$TR=T \$P2=P2-PCHAM	GAS 0790
TF=T-460. \$PRINT 810,TF	GAS 0800
810 FORMAT(* TEMPERATURE, DEGREES F **G12.5)	GAS 0810
QPERG=Q/(453.6*WLB) \$PRINT 814,QPERG	GAS 0812
814 FORMAT(* ENERGY RELEASE(KCAL/G) **G12.5)	GAS 0814
PRINT 830,G	GAS 0820
830 FORMAT(* SPECIFIC HEAT RATIO **G12.5)	GAS 0830
PRINT 870,P2	GAS 0860
870 FORMAT(* GAS OVERPRESSURE(PSI) **G12.5)	GAS 0870
RETURN	GAS 0880
END	GAS 0890

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SUBROUTINE TNT(L)
C POSITIVE-PHASE PROPERTIES FOR 1 LB TNT IN SEA-LEVEL AIR.
COMMON/TNTIN/RTNT,KMAX1
COMMON/TNTOUT/PTNT,TSTNT,TPTNT,TCTNT,PCNT,PSI(40),T1(40),T2(40)
1,JJ,XIMP1
DIMENSION P(108),R(108),TS(108),TP(108),TC(23)
C RADIAL DISTANCE FROM CHARGE CENTER (CM).
DATA R/ 4.054, 4.680, 5.700, 7.120, 9.200, 10.6, 12.4,
1 14.7, 17.6, 19.0, 21.7, 25.0, 28.1, 30.0, 32.0,
2 34.5, 37.4, 41.0, 46.0, 48.8, 52.4, 57.3, 65.,
3 66.4, 72.0, 79.0, 85.3, 89.0, 93.0, 98.4, 105.,
4 114., 125., 133., 142., 154., 171., 180., 196.,
5 218., 230., 251., 269., 289., 316., 350., 402.,
6 438., 483., 546., 640., 688., 781., 920., 1070.,
7 1160., 1260., 1400., 1580., 1830., 2200., 2450., 2790.,
8 3220., 3900., 4250., 4960., 6010., 7030., 7710., 8540.,
9 9600.,1.10E+4,1.29E+4,1.57E+4,1.77E+4,2.04E+4,2.40E+4,2.92E+4,
13.21E+4,3.79E+4,4.64E+4,5.48E+4,6.04E+4,6.73E+4,7.60E+4,8.76E+4,
1103600.,127400.,144200.,166400.,197100.,242700.,267800.,317600.,
2391300.,464300.,512500.,572500.,649100.,750400.,891100.,1.10E+6,
31.248E6,1.444E6,1.716E6,2.120E6,2.342E6/
C INCIDENT OVERPRESSURE (PSI).
DATA P/ 7800., 7000., 6000., 5000., 4000., 3500., 3000.,
1 2500., 2000., 1800., 1500., 1200., 1000., 900., 800.,
2 700., 600., 500., 400., 350., 300., 250., 190.,
3 180., 150., 120., 100., 90., 80., 70., 60.,
4 50., 40., 35., 30., 25., 20., 18., 15.,
5 12., 10., 9.0, 8.0, 7.0, 6.0, 5.0, 4.0,
6 3.5, 3.0, 2.5, 2.0, 1.8, 1.5, 1.2, 1.0,
7 .90, .80, .70, .60, .50, .40, .35, .30,
8 .25, .20, .18, .15, .12, .10, .09, .08,
9 .07, .06, .05, .04, .035, .03, .025, .02,
1 .018, .015, .012, .010, .009, .008, .007, .006,
1 .005, .004, .0035, .003, .0025, .002, .0018, .0015,
2 .0012, .0010, .0009, .0008, .0007, .0006, .0005, .0004,
3 .00035, .0003, .00025, .0002, .00018/
C SHOCK FRONT ARRIVAL TIME (SEC).
DATA TS/ 1.E-10, .78E-6,2.25E-6,4.52E-6,8.30E-6,10.9E-6,14.8E-6,
13.0E-6,17.4E-6,31.3E-6,39.3E-6,49.9E-6,60.7E-6,68.3E-6,76.2E-6,
288.8E-6,100.E-6,113.E-6,146.E-6,163.E-6,186.E-6,220.E-6,281.E-6,
3293.E-6,343.E-6,410.E-6,479.E-6,517.E-6,564.E-6,629.E-6,714.E-6,
4840.E-6,1.01E-3,1.14E-3,1.30E-3,1.51E-3,1.84E-3,2.02E-3,2.36E-3,
52.84E-3,3.32E-3,3.58E-3,4.04E-3,4.52E-3,5.19E-3,6.06E-3,7.40E-3,
68.38E-3,7.59E-3,11.3E-3,13.9E-3,15.2E-3,17.8E-3,21.7E-3,26.0E-3,
7 .0286, .0315, .0355, .0407, .0479, .0587, .0659, .0758,
8 .0884, .1082, .1185, .1392, .1700, .1998, .2198, .2438,
9 .2746, .3157, .3714, .4536, .5123, .5916, .6973, .8500,
1 .9352, 1.106, 1.355, 1.602, 1.766, 1.969, 2.225, 2.566,
1 3.036, 3.735, 4.229, 4.881, 5.783, 7.123, 7.861, 9.324,
2 11.49, 13.64, 15.05, 16.81, 19.06, 22.04, 26.18, 32.32,
3 36.66, 42.42, 50.42, 62.29, 68.81/
C DURATION OF POSITIVE OVERPRESSURE (SEC).
C FIRST 23 VALUES ARE DURATION IF CONTACT SURFACE HAD NOT ARRIVED.
DATA TP/3.90E-5,4.30E-5,5.00E-5,5.90E-5,7.20E-5,8.00E-5,9.00E-5,
11.02E-4,1.19E-4,1.28E-4,1.41E-4,1.60E-4,1.76E-4,1.86E-4,1.98E-4,

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22.10E-4,2.27E-4,2.45E-4,2.74E-4,2.89E-4,3.08E-4,3.35E-4,3.80E-4, TNT 0560
33.88E-4,4.20E-4,4.70E-4, .50E-3, .52E-3, .55E-3, .58E-3, .62E-3, TNT 0570
4 .68E-3, .76E-3, .83E-3, .90E-3, .99E-3,1.12E-3,1.19E-3,1.29E-3, TNT 0580
51.41E-3,1.51E-3,1.57E-3,1.65E-3,1.78E-3,1.84E-3,1.96E-3,2.13E-3, TNT 0590
62.24E-3,2.36E-3,2.52E-3,2.72E-3,2.82E-3,3.00E-3,3.17E-3,3.32E-3, TNT 0600
73.39E-3,3.45E-3,3.54E-3,3.62E-3,3.72E-3,3.84E-3,3.90E-3,3.97E-3, TNT 0610
84.05E-3,4.15E-3,4.19E-3,4.25E-3,4.34E-3,4.40E-3,4.45E-3,4.49E-3, TNT 0620
94.52E-3, .0., .0., .0., .0., .0., .0., .0., .0., TNT 0630
1 0.,0.,0.,0.,0.,0.,0.,0.,0.,0.,0.,0.,0.,0., TNT 0640
1 0.,0.,0.,0.,0.,0.,0.,0.,0.,0.,0.,0.,0.,0., TNT 0650
C TIME (SEC) BETWEEN SHOCK FRONT AND CONTACT SURFACE ARRIVAL. TNT 0660
DATA TC/ 1.E-10, .1E-6, .24E-6, .46E-6, .80E-6,1.10E-6, 1.5E-6, TNT 0670
12.24E-6, 3.2E-6,3.95E-6, 5.2E-6, 7.5E-6,10.9E-6,13.2E-6,15.9E-6, TNT 0680
220.2E-6,26.2E-6, 34.E-6, 49.E-6, 58.E-6, 77.E-6,116.E-6,452.E-6/ TNT 0690
C TNT 0740
C T1=TIME, T2=TIME SINCE SHOCK ARRIVAL. TNT 0750
FUN1(X1,X2)=EXP(ALOG(X1)+FRAC*ALOG(X2/X1)) TNT 0760
XIMPI=TCTNT=0. TNT 0770
C FIND DESIRED POINT IN TABLES. TNT 0780
DO 810 JJ=2,108 TNT 0790
IF(RTNT.LT.R(JJ)) GO TO 830 TNT 0800
810 CONTINUE $JJ=108 TNT 0810
C DESIRED DATA LIES BETWEEN R(JJ-1) AND R(JJ). TNT 0820
830 FRAC=ALOG(RTNT/R(JJ-1))/ALOG(R(JJ)/R(JJ-1)) TNT 0830
PTNT=FUN1(P(JJ-1),P(JJ)) $PSI(1)=PTNT TNT 0840
C CALC ONLY PSI(1) IF L=0 . TNT 0850
IF(L.EQ.0) RETURN TNT 0860
TSTNT=FUN1(TS(JJ-1),TS(JJ)) TNT 0870
TPTNT=FUN1(TP(JJ-1),TP(JJ)) TNT 0880
IF(JJ.LE.23) TCTNT=FUN1(TC(JJ-1),TC(JJ)) TNT 0890
C FIND SHAPE PARAMETER SIGMA. TNT 0940
SIGMA=228./RTNT-.95 TNT 0950
IF(JJ.LE.23) SIGMA=228./65.-.95 TNT 0955
T1(1)=TSTNT $T2(1)=0. TNT 0960
PSI(KMAX1)=0. $T1(KMAX1)=TSTNT+TPTNT $T2(KMAX1)=TPTNT TNT 0970
DO 1030 K=2,KMAX1 $IF(K.EQ.KMAX1) GO TO 1020 TNT 0980
TAU=FLOAT(K)/FLOAT(KMAX1) TNT 0990
PI=(1.-TAU)*EXP(-TAU*(1.+SIGMA/(.5+TAU))) TNT 1000
PSI(K)=PTNT*PI $T2(K)=TAU*TPTNT $T1(K)=TSTNT+T2(K) TNT 1010
1020 XIMPI=XIMPI+.5*(PSI(K)+PSI(K-1))*(T2(K)-T2(K-1)) TNT 1020
1030 CONTINUE $RETURN TNT 1030
END TNT 1330

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C	SUBROUTINE ARDC(ALTFT,PAMB,TAMB)	ARDC0010
C	ALTFT=ALTITUDE (KILOFEET).	ARDC0020
C	PAMB=AMBIENT PRESSURE (PSI).	ARDC0030
C	TAMB=AMBIENT TEMPERATURE (C).	ARDC0040
		ARDC0050
	ALTZ=ALTFT*.3048E3 SALTH=6356766.0*ALTZ/(6356766.0+ALTZ)	ARDC0060
	IF(ALTZ.GT.11000.) GO TO 100	ARDC0070
	TEMP=288.16-0.0065*ALTZ	ARDC0080
	PAMB=14.696178/(288.160/(288.160-0.0065*ALTZ))**5.25612218\$GOTO 400	ARDC0090
100	IF(ALTZ.GT.25000.) GO TO 130	ARDC0100
	TEMP=216.66	ARDC0110
	PAMB=3.28254528/(10.**(0.068483253E-3*(ALTZ-11000.0))) SGOTO 400	ARDC0120
130	IF(ALTZ.GT.47000.) GO TO 170	ARDC0130
	TEMP=216.66+0.003*(ALTZ-25000.0)	ARDC0140
	PAMB=0.36094654/((141.660+3.0E-3*ALTZ)/216.66)**11.38826473	ARDC0150
	GO TO 400	ARDC0160
170	IF(ALTZ.GT.53000.) GO TO 200	ARDC0170
	TEMP=282.66	ARDC0180
	PAMB=0.0174686/(10.**(0.0524926823E-3*(ALTZ-47000.0))) SGOTO 400	ARDC0190
200	IF(ALTZ.GT.79000.) GO TO 230	ARDC0200
	TEMP=282.66-0.0045*(ALTZ-53000.0)	ARDC0210
	PAMB=8.40408E-3/((282.66/TEMP)**7.592176) SGOTO 400	ARDC0220
230	IF(ALTZ.GT.90000.) GO TO 260	ARDC0230
	TEMP=165.66	ARDC0240
	PAMB=1.46198E-4*EXP (-0.0341647942*(ALTZ-79000.0)/165.66)\$GOTO 400	ARDC0250
260	IF(ALTZ.GT.105000.) GO TO 290	ARDC0260
	TEMP=165.66+0.0040*(ALTZ-90000.0)	ARDC0270
	PAMB=1.5519E-5*(165.66/TEMP)**8.541198 SGOTO 400	ARDC0280
290	IF(ALTZ.GT.160000.) GO TO 320	ARDC0290
	TEMP=225.66+0.02*(ALTZ-105000.0)	ARDC0300
	PAMB=1.04442E-6*(225.66/TEMP)**1.708239 SGOTO 400	ARDC0310
320	IF(ALTZ.GT.170000.) GO TO 350	ARDC0320
	TEMP=1325.66+0.01*(ALTZ-160000.0)	ARDC0330
	PAMB=5.14015E-8*(1325.66/TEMP)**3.4164794 SGOTO 400	ARDC0340
350	IF(ALTZ.GT.200000.) GO TO 380	ARDC0350
	TEMP=1425.66+0.005*(ALTZ-170000.0)	ARDC0360
	PAMB=4.0654E-8*(1425.66/TEMP)**6.832958 SGOTO 400	ARDC0370
380	TEMP=1575.66+0.0035*(ALTZ-200000.0)	ARDC0380
	PAMB=2.0595E-8*(1575.66/TEMP)**9.761369	ARDC0390
400	TAMB=TEMP-273.16 \$RETURN	ARDC0400
	END	ARDC0410