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TRANSONIC WIND TUNNEL TESTS ON A MISSILE
MODEL WITH CANARD WINGS. VOLUME II

R. E. de Kuyper

Calspan Corporation

Prepared for:

Army Missile Command

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TECHNICAL REPORT

AEROSCIENCES DIVISION 8-FOOT TRANSONIC WIND TUNNEL

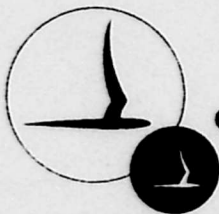
*TRANSONIC WIND TUNNEL TESTS ON A
MISSILE MODEL WITH CANARD WINGS*

Volume II of III
W.A. T17-093

R.E. de Kuyper
Report No. AA-4017-W-9

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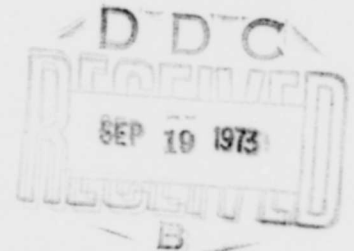
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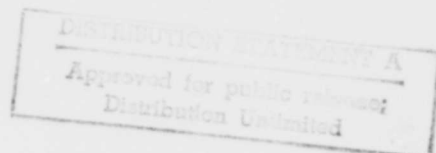
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TABLE VA
 BALANCE CAVITY AND BASE PRESSURE COEFFICIENTS, AND INCREMENTAL
 AXIAL FORCE COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	M _c	α	φ	C _{Pc}	C _{Pb1}	C _{Pb2}	ΔC _{Ac}	ΔC _{Ab}	C _{Au}
621	1.250	12.77	-0.00	-0.294	-0.282	-0.309	-0.10607	-0.18936	0.63166
621	1.251	0.02	-0.00	-0.235	-0.240	-0.250	-0.08469	-0.15723	0.53065
622	1.250	-3.04	-0.00	-0.234	-0.236	-0.253	-0.08442	-0.15681	0.52867
622	1.251	0.05	-0.00	-0.240	-0.245	-0.255	-0.08646	-0.16039	0.54979
622	1.250	1.13	-0.00	-0.238	-0.247	-0.257	-0.08648	-0.16144	0.55731
622	1.251	6.43	-0.00	-0.245	-0.242	-0.263	-0.08569	-0.16194	0.56511
622	1.250	9.58	-0.00	-0.266	-0.237	-0.266	-0.08842	-0.15955	0.59088
622	1.251	12.80	-0.00	-0.293	-0.280	-0.282	-0.09606	-0.16966	0.61959
622	1.251	0.05	-0.00	-0.238	-0.244	-0.252	-0.10571	-0.18848	0.65619
623	1.252	-2.99	-0.00	-0.225	-0.232	-0.241	-0.08105	-0.15165	0.56218
623	1.252	1.19	-0.00	-0.228	-0.238	-0.242	-0.08216	-0.15403	0.59076
623	1.251	3.30	-0.00	-0.232	-0.245	-0.250	-0.08373	-0.15863	0.60405
623	1.251	6.47	-0.00	-0.224	-0.235	-0.249	-0.08072	-0.15477	0.60732
623	1.253	9.61	-0.00	-0.251	-0.239	-0.258	-0.08303	-0.16279	0.67038
623	1.252	12.85	-0.00	-0.278	-0.270	-0.298	-0.10021	-0.18219	0.71038
623	1.252	0.10	-0.00	-0.229	-0.240	-0.245	-0.08260	-0.15553	0.58787
624	0.901	-2.96	-0.00	-0.104	-0.111	-0.137	-0.03746	-0.07984	0.32593
624	0.902	1.10	-0.00	-0.108	-0.139	-0.140	-0.03910	-0.08952	0.36150
624	0.900	3.21	-0.00	-0.111	-0.137	-0.130	-0.04277	-0.08578	0.37194
624	0.900	6.39	-0.00	-0.122	-0.117	-0.142	-0.04421	-0.08935	0.37985
624	0.898	9.56	-0.00	-0.134	-0.130	-0.172	-0.04854	-0.09707	0.39783
624	0.899	12.87	-0.00	-0.170	-0.157	-0.216	-0.06136	-0.11969	0.48713
624	0.899	0.07	-0.00	-0.107	-0.138	-0.141	-0.03866	-0.08976	0.36073
625	0.901	-3.03	-0.00	-0.103	-0.109	-0.131	-0.03732	-0.07714	0.28910
625	0.900	1.08	-0.00	-0.104	-0.131	-0.131	-0.03753	-0.08415	0.31452
625	0.899	3.19	-0.00	-0.109	-0.133	-0.126	-0.03948	-0.08308	0.31019
625	0.900	6.39	-0.00	-0.126	-0.131	-0.143	-0.04321	-0.08793	0.33524
625	0.901	9.56	-0.00	-0.136	-0.130	-0.153	-0.04570	-0.08808	0.35695
625	0.900	12.86	-0.00	-0.170	-0.159	-0.174	-0.04921	-0.09785	0.39793
625	0.900	0.01	-0.00	-0.107	-0.133	-0.133	-0.03860	-0.08533	0.31224
626	0.900	-3.05	-0.00	-0.102	-0.107	-0.125	-0.03689	-0.07475	0.27834
626	0.900	1.00	-0.00	-0.105	-0.129	-0.128	-0.03807	-0.08268	0.29569
626	0.901	3.06	-0.00	-0.108	-0.129	-0.124	-0.03911	-0.08145	0.29811

TABLE VA
 BALANCE CAVITY AND BASE PRESSURE COEFFICIENTS, AND INCREMENTAL
 AXIAL FORCE COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	M _c	α	φ	C _{Pc}	C _{Pb1}	C _{Pb2}	ΔC _{Ac}	ΔC _{Ab}	C _{Au}
626	0.900	3.15	-0.00	-0.116	-0.124	-0.136	-0.04194	-0.08380	0.31752
626	0.901	6.28	-0.00	-0.126	-0.121	-0.152	-0.04544	-0.06704	0.33201
626	0.900	9.38	-0.00	-0.135	-0.130	-0.173	-0.04888	-0.09717	0.33627
626	0.900	12.56	-0.00	-0.144	-0.162	-0.217	-0.06290	-0.12179	0.37145
626	0.900	0.00	-0.00	-0.105	-0.128	-0.128	-0.03800	-0.08224	0.29531
627	0.900	-3.06	-0.00	-0.104	-0.107	-0.126	-0.03756	-0.07496	0.27860
627	0.899	-0.02	-0.00	-0.105	-0.122	-0.126	-0.03790	-0.07971	0.28554
627	0.900	1.01	-0.00	-0.108	-0.122	-0.124	-0.03907	-0.07915	0.28760
627	0.900	3.11	-0.00	-0.114	-0.121	-0.132	-0.04120	-0.08133	0.30004
627	0.901	6.25	-0.00	-0.123	-0.121	-0.151	-0.0446	-0.08747	0.31254
627	0.899	9.35	-0.00	-0.137	-0.130	-0.176	-0.04956	-0.09832	0.31554
627	0.899	12.54	-0.00	-0.175	-0.165	-0.219	-0.06330	-0.12316	0.35346
627	0.901	-0.02	-0.00	-0.104	-0.122	-0.124	-0.03760	-0.07913	0.28449
628	0.899	3.09	-0.00	-0.106	-0.108	-0.124	-0.03849	-0.07448	0.28072
628	0.898	-0.04	-0.00	-0.107	-0.121	-0.125	-0.03866	-0.07906	0.28017
628	0.900	1.01	-0.00	-0.106	-0.121	-0.122	-0.03837	-0.07829	0.28341
628	0.900	3.10	-0.00	-0.112	-0.120	-0.131	-0.04036	-0.08066	0.29190
628	0.901	6.23	-0.00	-0.120	-0.121	-0.150	-0.04346	-0.08686	0.29974
628	0.901	9.34	-0.00	-0.137	-0.133	-0.177	-0.04965	-0.09950	0.30551
628	0.898	12.53	-0.00	-0.181	-0.169	-0.220	-0.06516	-0.12506	0.33970
628	0.901	-0.04	-0.00	-0.106	-0.119	-0.124	-0.03834	-0.07812	0.28179
629	0.900	3.10	-0.00	-0.107	-0.107	-0.122	-0.03886	-0.07388	0.28407
629	0.898	-0.06	-0.00	-0.107	-0.121	-0.127	-0.03856	-0.07954	0.28095
629	0.900	1.07	-0.00	-0.107	-0.121	-0.120	-0.03870	-0.07752	0.28342
629	0.899	3.07	-0.00	-0.112	-0.121	-0.127	-0.04037	-0.07943	0.28911
629	0.899	6.22	-0.00	-0.124	-0.126	-0.154	-0.04494	-0.08982	0.29402
629	0.897	9.35	-0.00	-0.144	-0.136	-0.180	-0.05191	-0.10158	0.29946
629	0.897	12.52	-0.00	-0.183	-0.170	-0.222	-0.06614	-0.12590	0.33446
629	0.898	-0.05	-0.00	-0.104	-0.117	-0.124	-0.03750	-0.07743	0.27867
630	0.898	3.11	-0.00	-0.116	-0.113	-0.129	-0.04189	-0.07787	0.29326
630	0.898	-0.07	-0.00	-0.107	-0.118	-0.128	-0.03795	-0.07907	0.28267
630	0.897	0.99	-0.00	-0.107	-0.123	-0.121	-0.03865	-0.07847	0.28175
630	0.898	3.05	-0.00	-0.110	-0.123	-0.127	-0.03974	-0.08013	0.28111
630	0.899	6.17	-0.00	-0.120	-0.123	-0.151	-0.04329	-0.08791	0.28582
630	0.899	9.28	-0.00	-0.142	-0.135	-0.181	-0.05135	-0.10166	0.28950
630	0.899	12.50	-0.00	-0.179	-0.168	-0.218	-0.06457	-0.12387	0.31973
630	0.899	-0.06	-0.00	-0.106	-0.119	-0.127	-0.03841	-0.07907	0.28251

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 BALANCE CAVITY AND BASE PRESSURE COEFFICIENTS, AND INCREMENTAL
 AXIAL FORCE COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	M _c	α	φ	C _p c	C _p b ₁	C _p b ₂	ΔC _A c	ΔC _A b	C _A u
632	1.252	-3.16	-0.00	-0.214	-0.214	-0.250	-0.07730	-0.14256	0.51482
632	1.251	-1.11	-0.00	-0.212	-0.216	-0.229	-0.07635	-0.14327	0.50574
632	1.255	-0.05	-0.00	-0.211	-0.220	-0.230	-0.07612	-0.14367	0.50276
632	1.252	0.47	-0.00	-0.213	-0.221	-0.233	-0.07621	-0.14435	0.50210
632	1.252	0.99	-0.00	-0.213	-0.224	-0.238	-0.07690	-0.14468	0.50364
632	1.252	3.12	-0.00	-0.216	-0.223	-0.236	-0.07807	-0.14789	0.50426
632	1.253	6.39	-0.00	-0.236	-0.242	-0.254	-0.08513	-0.15378	0.52205
632	1.253	9.61	-0.00	-0.256	-0.272	-0.297	-0.09228	-0.16575	0.54423
632	1.251	12.06	-0.00	-0.280	-0.291	-0.330	-0.10085	-0.18251	0.56573
				-0.212	-0.221	-0.233	-0.07647	-0.14457	0.50094
633	1.251	3.12	-0.00	-0.219	-0.219	-0.234	-0.07897	-0.14533	0.50763
633	1.252	-1.08	-0.00	-0.216	-0.220	-0.232	-0.07779	-0.14491	0.50111
633	1.255	-0.04	-0.00	-0.213	-0.221	-0.231	-0.07689	-0.14489	0.49903
633	1.254	0.48	-0.00	-0.215	-0.224	-0.233	-0.07756	-0.14653	0.49888
633	1.250	1.13	-0.00	-0.219	-0.226	-0.242	-0.07828	-0.14832	0.50196
633	1.251	3.26	-0.00	-0.238	-0.242	-0.257	-0.08582	-0.15515	0.52431
633	1.253	9.42	-0.00	-0.262	-0.277	-0.300	-0.09455	-0.16890	0.54802
633	1.252	12.06	-0.00	-0.285	-0.294	-0.333	-0.10265	-0.18494	0.57265
				-0.217	-0.224	-0.233	-0.07813	-0.14659	0.49765
634	1.252	3.13	-0.00	-0.219	-0.219	-0.236	-0.07915	-0.14579	0.50493
634	1.251	-1.05	-0.00	-0.217	-0.222	-0.233	-0.07837	-0.14571	0.49910
634	1.253	-0.52	-0.00	-0.218	-0.224	-0.234	-0.07855	-0.14675	0.49891
634	1.253	1.00	-0.00	-0.218	-0.226	-0.237	-0.07868	-0.14829	0.49974
634	1.253	3.17	-0.00	-0.219	-0.226	-0.237	-0.07902	-0.14854	0.50226
634	1.253	6.42	-0.00	-0.239	-0.247	-0.258	-0.08627	-0.14964	0.50691
634	1.253	9.42	-0.00	-0.263	-0.276	-0.279	-0.09503	-0.15520	0.52601
634	1.252	12.05	-0.00	-0.285	-0.298	-0.300	-0.10295	-0.16862	0.55176
				-0.218	-0.224	-0.235	-0.07876	-0.14736	0.49772
635	1.250	3.13	-0.00	-0.220	-0.222	-0.237	-0.07927	-0.14710	0.4985
635	1.251	-1.05	-0.00	-0.218	-0.227	-0.234	-0.07860	-0.14630	0.50310
635	1.251	0.51	-0.00	-0.218	-0.229	-0.236	-0.07835	-0.14813	0.50395
635	1.250	1.00	-0.00	-0.219	-0.229	-0.241	-0.07918	-0.15068	0.50628
635	1.251	3.14	-0.00	-0.222	-0.230	-0.247	-0.08000	-0.15288	0.51344
635	1.251	6.48	-0.00	-0.238	-0.247	-0.260	-0.08574	-0.15615	0.53171
635	1.251	9.43	-0.00	-0.267	-0.277	-0.279	-0.09478	-0.16678	0.55864
635	1.251	12.05	-0.00	-0.287	-0.291	-0.303	-0.10357	-0.18710	0.5854

TABLE VA
 BALANCE CAVITY AND BASE PRESSURE COEFFICIENTS, AND INCREMENTAL
 AXIAL FORCE COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	M _c	α	φ	C _{Pc}	C _{Pb1}	C _{Pb2}	ΔC _{Ac}	ΔC _{Ab}	C _{Au}
635	17	1.252	-0.03	-0.00	-0.217	-0.226	-0.236	-0.07828	-0.14799	0.50090
636	1	1.251	3.11	0.00	-0.220	-0.223	-0.239	-0.07942	-0.14819	0.50452
636	2	1.251	-1.03	0.00	-0.221	-0.227	-0.238	-0.07939	-0.14895	0.50545
636	3	1.251	0.01	0.00	-0.221	-0.231	-0.240	-0.07991	-0.15110	0.50769
636	4	1.252	0.54	0.00	-0.221	-0.231	-0.244	-0.07965	-0.15109	0.50800
636	5	1.251	1.03	0.00	-0.223	-0.233	-0.250	-0.08054	-0.15293	0.51131
636	6	1.251	3.15	0.00	-0.226	-0.233	-0.262	-0.08153	-0.15504	0.52104
636	7	1.249	6.31	0.00	-0.238	-0.229	-0.281	-0.08600	-0.15715	0.53956
636	8	1.252	9.43	0.00	-0.263	-0.248	-0.306	-0.09476	-0.16968	0.56613
636	9	1.249	12.66	0.00	-0.288	-0.282	-0.306	-0.10401	-0.18853	0.59512
636	10	1.250	-0.02	-0.00	-0.222	-0.231	-0.240	-0.08013	-0.15100	0.50679
637	1	1.251	3.08	0.00	-0.223	-0.227	-0.244	-0.08055	-0.15132	0.50959
637	2	1.252	-1.00	0.00	-0.225	-0.234	-0.243	-0.08119	-0.15306	0.51694
637	3	1.251	0.00	0.00	-0.227	-0.238	-0.244	-0.08192	-0.15451	0.52245
637	4	1.251	0.56	0.00	-0.226	-0.236	-0.244	-0.08119	-0.15497	0.52220
637	5	1.249	1.08	0.00	-0.226	-0.237	-0.246	-0.08162	-0.15750	0.53517
637	6	1.250	3.12	0.00	-0.231	-0.237	-0.254	-0.08353	-0.15825	0.55242
637	7	1.252	6.32	0.00	-0.241	-0.230	-0.263	-0.08602	-0.16873	0.57842
637	8	1.249	9.44	0.00	-0.263	-0.245	-0.281	-0.09476	-0.18790	0.60771
637	9	1.253	12.67	0.00	-0.289	-0.268	-0.305	-0.10422	-0.18790	0.62069
637	10	1.250	0.01	-0.00	-0.228	-0.238	-0.246	-0.08231	-0.15540	0.52069
638	1	1.251	3.07	0.00	-0.228	-0.239	-0.249	-0.08219	-0.15437	0.52093
638	2	1.252	-0.98	0.00	-0.229	-0.239	-0.247	-0.08262	-0.15582	0.53140
638	3	1.250	0.03	0.00	-0.233	-0.243	-0.249	-0.08412	-0.15799	0.53574
638	4	1.251	0.57	0.00	-0.231	-0.243	-0.249	-0.08328	-0.15751	0.53789
638	5	1.252	1.10	0.00	-0.229	-0.241	-0.256	-0.08279	-0.15712	0.53766
638	6	1.251	3.22	0.00	-0.233	-0.239	-0.264	-0.08421	-0.15900	0.54962
638	7	1.250	6.35	0.00	-0.239	-0.231	-0.281	-0.08624	-0.16917	0.59131
638	8	1.252	9.45	0.00	-0.262	-0.247	-0.305	-0.09431	-0.18767	0.62260
638	9	1.252	12.68	0.00	-0.289	-0.280	-0.305	-0.10431	-0.18767	0.63470
638	10	1.250	0.03	-0.00	-0.232	-0.242	-0.249	-0.08381	-0.15774	0.53470
639	1	1.251	3.02	0.00	-0.231	-0.237	-0.250	-0.08350	-0.15609	0.53435
639	2	1.250	-0.95	0.00	-0.233	-0.243	-0.249	-0.0839A	-0.15697	0.54503
639	3	1.251	0.05	0.00	-0.234	-0.245	-0.249	-0.08355	-0.15769	0.55267
639	4	1.251	0.59	0.00	-0.236	-0.246	-0.252	-0.08520	-0.15969	0.55469
639	5	1.251	1.12	0.00	-0.235	-0.246	-0.254	-0.08480	-0.16019	0.55689
639	6	1.251	3.25	0.00	-0.235	-0.242	-0.258	-0.08480	-0.16019	0.56421
639	7	1.250	6.37	0.00	-0.240	-0.231	-0.264	-0.08659	-0.16894	0.58101

TABLE VA
 BALANCE CAVITY AND BASE PRESSURE COEFFICIENTS, AND INCREMENTAL
 AXIAL FORCE COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	M _c	α	Φ	C _{Pc}	C _{Pb1}	C _{Pb2}	ΔC _{Ac}	ΔC _{Ab}	C _{Au}
639	1.252	9.46	-0.00	-0.262	-0.244	-0.261	-0.09435	-0.16824	0.60587
639	1.252	12.69	-0.00	-0.269	-0.260	-0.306	-0.10413	-0.18794	0.63781
639	1.251	0.05	-0.00	-0.234	-0.244	-0.250	-0.08459	-0.15825	0.54934
640	1.252	-3.01	-0.00	-0.229	0.235	-0.246	-0.08255	-0.15419	0.55073
640	1.252	-0.94	-0.00	-0.231	-0.240	-0.247	-0.08321	-0.15640	0.55412
640	1.251	0.61	-0.00	-0.233	-0.243	-0.248	-0.08408	-0.15733	0.57021
640	1.252	1.13	-0.00	-0.235	-0.246	-0.251	-0.08466	-0.15927	0.57463
640	1.250	3.27	-0.00	-0.236	-0.243	-0.254	-0.08517	-0.16062	0.57863
640	1.252	6.35	-0.00	-0.234	-0.243	-0.258	-0.08452	-0.16062	0.58024
640	1.250	9.50	-0.00	-0.237	-0.231	-0.261	-0.08555	-0.15777	0.60024
640	1.254	12.66	-0.00	-0.257	-0.273	-0.278	-0.09283	-0.16640	0.62613
640	1.252	0.06	-0.00	-0.232	-0.243	-0.301	-0.10162	-0.18412	0.65453
641	1.250	0.01	-0.00	-0.218	-0.229	-0.236	-0.07870	-0.14931	0.50475
641	1.253	0.00	-0.00	-0.222	-0.231	-0.239	-0.08018	-0.15054	0.51807
641	1.252	0.01	-0.00	-0.228	-0.235	-0.244	-0.08235	-0.15338	0.53311
642	1.252	-3.13	0.00	-0.225	0.229	-0.246	-0.08106	-0.15247	0.51863
642	1.252	-0.04	0.00	-0.220	-0.233	-0.240	-0.07951	-0.15178	0.51216
642	1.251	1.04	0.00	-0.221	-0.233	-0.237	-0.07963	-0.15072	0.51099
642	1.251	2.09	0.00	-0.223	-0.232	-0.242	-0.07956	-0.15047	0.51218
642	1.251	3.09	-0.00	-0.223	-0.233	-0.243	-0.08014	-0.15277	0.51462
642	1.252	4.19	-0.00	-0.226	-0.231	-0.245	-0.08070	-0.15271	0.51900
642	1.252	-0.06	0.00	-0.221	-0.232	-0.239	-0.07959	-0.15085	0.51124
643	1.252	-3.12	-0.00	-0.219	0.227	-0.236	-0.07898	-0.14830	0.51300
643	1.251	-0.05	-0.00	-0.216	-0.228	-0.234	-0.07797	-0.14842	0.50604
643	1.252	1.43	-0.00	-0.218	-0.230	-0.236	-0.07863	-0.14944	0.50537
643	1.251	2.05	-0.00	-0.219	-0.229	-0.235	-0.07817	-0.14861	0.50707
643	1.250	3.11	-0.00	-0.222	-0.232	-0.238	-0.07901	-0.15003	0.51048
643	1.251	4.16	-0.00	-0.225	-0.228	-0.245	-0.08126	-0.15151	0.51792
643	1.251	-0.05	-0.00	-0.218	-0.230	-0.236	-0.07861	-0.14937	0.50427
644	1.252	-3.13	-0.00	-0.218	0.223	-0.236	-0.07857	-0.14702	0.50696
644	1.253	-0.06	-0.00	-0.218	-0.229	-0.237	-0.07868	-0.14929	0.50285
644	1.253	1.04	-0.00	-0.215	-0.227	-0.236	-0.07771	-0.14848	0.50380
644	1.250	2.05	-0.00	-0.219	-0.230	-0.241	-0.07873	-0.15118	0.50384
644	1.250	3.05	-0.00	-0.219	-0.230	-0.243	-0.07915	-0.15118	0.50891

TABLE VA
BALANCE CAVITY AND BASE PRESSURE COEFFICIENTS, AND INCREMENTAL AXIAL FORCE COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	M _c	α	β	C _{Pc}	C _{Pb1}	C _{Pb2}	ΔC _{Ac}	ΔC _{Ab}	C _{Au}
644	1.252	3.11	-0.00	-0.219	-0.229	-0.247	-0.07917	-0.15272	0.51154
644	1.252	4.16	-0.00	-0.222	-0.227	-0.252	-0.08015	-0.15352	0.51500
644	1.252	-0.03	-0.00	-0.217	-0.228	-0.236	-0.07812	-0.14852	0.50228
645	1.253	-3.15	-0.00	-0.217	-0.221	-0.235	-0.07823	-0.14623	0.50785
645	1.253	-0.03	-0.00	-0.214	-0.226	-0.235	-0.07735	-0.14797	0.50204
645	1.253	0.51	-0.00	-0.216	-0.228	-0.238	-0.07806	-0.14940	0.50334
645	1.250	1.02	-0.00	-0.217	-0.230	-0.241	-0.07817	-0.15102	0.50562
645	1.252	2.10	-0.00	-0.218	-0.229	-0.243	-0.07850	-0.15134	0.50918
645	1.251	3.19	-0.00	-0.219	-0.229	-0.247	-0.07899	-0.15284	0.51125
645	1.251	4.19	-0.00	-0.223	-0.228	-0.252	-0.08040	-0.15404	0.51470
645	1.251	-0.02	-0.00	-0.215	-0.226	-0.236	-0.07771	-0.14808	0.50172
646	1.251	3.12	-0.00	-0.218	-0.223	-0.237	-0.07870	-0.14761	0.50883
646	1.251	-0.02	-0.00	-0.218	-0.229	-0.239	-0.07851	-0.15013	0.50414
646	1.252	0.49	-0.00	-0.217	-0.229	-0.240	-0.07843	-0.15036	0.50434
646	1.251	1.02	-0.00	-0.218	-0.230	-0.242	-0.07872	-0.15150	0.50639
646	1.250	2.05	-0.00	-0.220	-0.232	-0.248	-0.07944	-0.15494	0.50942
646	1.251	3.15	-0.00	-0.221	-0.230	-0.252	-0.07958	-0.15454	0.51209
646	1.251	4.16	-0.00	-0.224	-0.228	-0.256	-0.08085	-0.15513	0.51540
646	1.252	-0.03	-0.00	-0.217	-0.228	-0.238	-0.07827	-0.14925	0.50292
647	1.251	3.13	-0.00	-0.219	-0.224	-0.236	-0.07888	-0.14745	0.50869
647	1.250	-0.02	-0.00	-0.219	-0.229	-0.240	-0.07897	-0.15033	0.50571
647	1.250	0.48	-0.00	-0.219	-0.230	-0.240	-0.07885	-0.15064	0.50695
647	1.250	1.04	-0.00	-0.220	-0.232	-0.243	-0.07946	-0.15248	0.51054
647	1.252	2.08	-0.00	-0.219	-0.229	-0.249	-0.07911	-0.15359	0.51305
647	1.251	3.14	-0.00	-0.225	-0.229	-0.256	-0.08121	-0.15545	0.51662
647	1.252	4.17	-0.00	-0.225	-0.228	-0.237	-0.07854	-0.14907	0.50275
647	1.252	-0.05	-0.00	-0.217	-0.228	-0.237	-0.07854	-0.14907	0.50275
648	1.249	3.14	-0.00	-0.219	-0.223	-0.236	-0.07891	-0.14712	0.50830
648	1.252	-0.04	-0.00	-0.217	-0.226	-0.237	-0.07815	-0.14836	0.50458
648	1.252	0.00	-0.00	-0.216	-0.229	-0.239	-0.07778	-0.14928	0.51076
648	1.251	0.99	-0.00	-0.219	-0.229	-0.245	-0.07910	-0.15208	0.51291
648	1.251	2.08	-0.00	-0.219	-0.230	-0.249	-0.07899	-0.15341	0.51481
648	1.251	3.13	-0.00	-0.220	-0.228	-0.254	-0.07933	-0.15451	0.51294
648	1.251	4.19	-0.00	-0.223	-0.227	-0.258	-0.08044	-0.15529	0.51553
648	1.250	-0.07	-0.00	-0.216	-0.226	-0.237	-0.07803	-0.14870	0.50358
649	1.252	3.13	-0.00	-0.219	-0.224	-0.233	-0.07914	-0.14651	0.51127
649	1.251	-0.04	-0.00	-0.218	-0.226	-0.237	-0.07859	-0.14860	0.50601

TABLE VA
BALANCE CAVITY AND BASE PRESSURE COEFFICIENTS, AND INCREMENTAL AXIAL FORCE COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	M _c	α	φ	C _p c	C _p b ₁	C _p b ₂	ΔC _A c	ΔC _A b	C _A u
649	3	1.251	0.51	-0.00	-0.220	-0.230	-0.242	-0.07924	-0.15117	0.50710
649	4	1.251	1.01	-0.00	-0.222	-0.232	-0.244	-0.07945	-0.15259	0.51009
649	5	1.250	3.09	-0.00	-0.222	-0.231	-0.250	-0.08014	-0.15497	0.51445
649	6	1.251	3.13	-0.00	-0.222	-0.228	-0.255	-0.08003	-0.15590	0.51576
649	7	1.251	4.18	-0.00	-0.222	-0.227	-0.258	-0.08027	-0.15591	0.51845
649	8	1.251	-0.04	-0.00	-0.218	-0.227	-0.237	-0.07867	-0.14897	0.50568
650	1	1.252	3.13	-0.00	-0.224	-0.237	-0.239	-0.08090	-0.15286	0.52326
650	2	1.252	-0.04	-0.00	-0.221	-0.229	-0.245	-0.07970	-0.15218	0.51747
650	3	1.253	1.02	-0.00	-0.219	-0.231	-0.250	-0.07923	-0.15407	0.51765
650	4	1.252	1.08	-0.00	-0.220	-0.232	-0.256	-0.07900	-0.15648	0.52024
650	5	1.252	3.14	-0.00	-0.221	-0.232	-0.265	-0.07955	-0.15990	0.52094
650	6	1.251	4.19	-0.00	-0.221	-0.228	-0.264	-0.07976	-0.15921	0.52054
650	7	1.251	-0.04	-0.00	-0.223	-0.230	-0.248	-0.07997	-0.15237	0.52033
650	8	1.250	-0.04	-0.00	-0.223	-0.230	-0.248	-0.08029	-0.15345	0.51616
651	1	1.250	3.14	-0.00	-0.228	-0.243	-0.237	-0.08240	-0.15383	0.52767
651	2	1.250	-0.03	-0.00	-0.229	-0.237	-0.248	-0.08249	-0.15556	0.52935
651	3	1.252	1.02	-0.00	-0.226	-0.236	-0.248	-0.08192	-0.15560	0.52977
651	4	1.252	1.08	-0.00	-0.226	-0.236	-0.253	-0.08147	-0.15762	0.52947
651	5	1.251	3.12	-0.00	-0.226	-0.241	-0.265	-0.08166	-0.16217	0.52816
651	6	1.251	4.19	-0.00	-0.226	-0.239	-0.256	-0.08144	-0.15857	0.52804
651	7	1.250	-0.05	-0.00	-0.224	-0.232	-0.248	-0.08072	-0.15857	0.52854
651	8	1.250	-0.05	-0.00	-0.228	-0.237	-0.247	-0.08229	-0.15522	0.52857
652	1	1.253	3.21	45.00	-0.227	-0.237	-0.245	-0.08180	-0.15465	0.52935
652	2	1.250	-0.03	45.00	-0.219	-0.225	-0.239	-0.07890	-0.14864	0.51319
652	3	1.250	3.04	45.00	-0.225	-0.235	-0.243	-0.08129	-0.15354	0.52893
652	4	1.252	-0.11	45.00	-0.217	-0.223	-0.235	-0.07815	-0.14677	0.51196
652	5	1.251	0.43	44.99	-0.218	-0.225	-0.237	-0.07848	-0.14825	0.51047
652	6	1.252	0.93	44.99	-0.218	-0.226	-0.240	-0.07870	-0.14927	0.50955
652	7	1.252	3.01	44.99	-0.217	-0.229	-0.239	-0.07812	-0.15004	0.50826
652	8	1.251	4.10	44.99	-0.219	-0.235	-0.242	-0.07910	-0.15284	0.50966
652	9	1.251	-0.09	44.99	-0.228	-0.252	-0.254	-0.08220	-0.16230	0.51198
652	10	1.252	-0.09	44.99	-0.219	-0.224	-0.236	-0.07891	-0.14756	0.51198
652	11	1.252	3.25	45.00	-0.218	-0.228	-0.236	-0.07882	-0.14879	0.50866
652	12	1.252	-0.05	44.99	-0.214	-0.223	-0.231	-0.07782	-0.14547	0.50512
652	13	1.251	0.98	44.99	-0.216	-0.225	-0.233	-0.07782	-0.14672	0.50172
652	14	1.250	3.03	44.99	-0.216	-0.225	-0.239	-0.07674	-0.15005	0.51091
652	15	1.250	6.23	44.99	-0.242	-0.274	-0.270	-0.08727	-0.17426	0.53535
652	16	1.250	9.40	44.99	-0.260	-0.293	-0.287	-0.09372	-0.18584	0.54735

TABLE VA
BALANCE CAVITY AND BASE PRESSURE COEFFICIENTS, AND INCREMENTAL AXIAL FORCE COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	M _c	α	φ	C _{pC}	C _{p_{b1}}	C _{p_{b2}}	ΔC _{A_c}	ΔC _{A_b}	C _{A_u}
653	1.252	12.56	44.99	-0.272	-0.309	-0.294	-0.098	0.16	0.559
653	1.250	-0.05	44.99	-0.215	-0.222	-0.231	-0.077	0.14	0.498
654	1.250	3.24	45.00	-0.230	-0.237	-0.246	-0.083	0.15	0.528
654	1.250	-0.09	44.99	-0.220	-0.225	-0.237	-0.079	0.14	0.511
654	1.251	3.05	44.99	-0.220	-0.235	-0.243	-0.079	0.14	0.508
654	1.250	6.19	44.99	-0.244	-0.274	-0.271	-0.087	0.17	0.527
654	1.250	9.36	44.99	-0.264	-0.298	-0.298	-0.095	0.19	0.547
654	1.251	12.57	44.99	-0.277	-0.313	-0.303	-0.099	0.19	0.547
654	1.251	-0.10	44.99	-0.220	-0.225	-0.237	-0.079	0.14	0.511
655	1.252	3.18	45.00	-0.219	-0.223	-0.238	-0.078	0.14	0.508
655	1.254	-0.02	44.99	-0.215	-0.223	-0.232	-0.077	0.14	0.501
655	1.253	0.98	44.99	-0.216	-0.226	-0.234	-0.077	0.14	0.503
655	1.251	3.09	44.99	-0.222	-0.232	-0.243	-0.080	0.15	0.514
655	1.250	6.25	44.99	-0.243	-0.272	-0.271	-0.082	0.17	0.532
655	1.251	9.43	44.99	-0.259	-0.292	-0.285	-0.093	0.18	0.549
655	1.250	12.61	44.99	-0.273	-0.310	-0.297	-0.098	0.19	0.564
655	1.251	-0.04	44.99	-0.215	-0.224	-0.232	-0.077	0.14	0.509
656	1.251	3.14	44.99	-0.220	-0.230	-0.240	-0.079	0.15	0.505
656	1.252	-0.00	44.99	-0.220	-0.229	-0.236	-0.079	0.14	0.510
656	1.251	1.01	44.99	-0.216	-0.226	-0.234	-0.077	0.14	0.511
656	1.250	3.28	44.99	-0.236	-0.271	-0.270	-0.086	0.17	0.527
656	1.251	6.44	44.99	-0.258	-0.292	-0.284	-0.092	0.18	0.546
656	1.251	9.61	44.99	-0.270	-0.307	-0.293	-0.097	0.19	0.562
656	1.252	12.80	44.99	-0.288	-0.328	-0.315	-0.103	0.20	0.576
657	1.252	3.11	44.99	-0.221	-0.230	-0.245	-0.079	0.15	0.517
657	1.252	-0.03	44.99	-0.221	-0.230	-0.245	-0.079	0.15	0.517
657	1.252	1.09	44.99	-0.225	-0.236	-0.243	-0.081	0.15	0.528
657	1.251	3.21	44.99	-0.231	-0.242	-0.248	-0.085	0.15	0.535
657	1.251	6.36	44.99	-0.236	-0.265	-0.267	-0.085	0.17	0.550
657	1.251	9.51	44.99	-0.251	-0.288	-0.275	-0.090	0.18	0.565
657	1.250	12.69	44.99	-0.265	-0.308	-0.288	-0.095	0.19	0.580
658	1.253	3.06	44.99	-0.227	-0.239	-0.243	-0.081	0.15	0.533
658	1.252	-0.11	44.99	-0.227	-0.232	-0.251	-0.081	0.15	0.535

TABLE VA
BALANCE CAVITY AND BASE PRESSURE COEFFICIENTS, AND INCREMENTAL
AXIAL FORCE COEFFICIENTS - AERODYNAMIC PHASE:

Run Pt.	M _c	α	φ	C _p c	C _p b1	C _p b2	ΔC _A c	ΔC _A b	C _A u
65A	3	1.252	44.99	-0.234	-0.247	-0.248	-0.08449	-0.15851	0.56802
65A	4	3.24	44.99	-0.233	-0.244	-0.250	-0.08391	-0.15823	0.57415
65A	5	6.38	44.99	-0.233	-0.261	-0.265	-0.08388	-0.16856	0.59062
65A	6	9.53	44.99	-0.249	-0.287	-0.273	-0.08970	-0.17975	0.60742
65A	7	12.71	44.99	-0.259	-0.307	-0.292	-0.09719	-0.19187	0.62458
65A	8	0.08	44.99	-0.237	-0.250	-0.251	-0.08552	-0.16070	0.56135
659	1	1.251	44.99	-0.230	-0.232	-0.255	-0.08301	-0.15634	0.58999
659	2	0.18	44.99	-0.242	-0.252	-0.254	-0.08717	-0.16209	0.62512
659	3	1.18	44.99	-0.241	-0.254	-0.255	-0.08701	-0.16298	0.63439
659	4	3.27	44.99	-0.235	-0.246	-0.249	-0.08472	-0.15860	0.64901
659	5	6.42	44.99	-0.234	-0.259	-0.267	-0.08430	-0.16852	0.66876
659	6	9.55	44.99	-0.251	-0.287	-0.275	-0.09059	-0.18009	0.68395
659	7	12.71	44.99	-0.269	-0.301	-0.291	-0.09717	-0.18964	0.68395
659	8	0.14	44.99	-0.240	-0.251	-0.253	-0.08659	-0.16165	0.62575
660	1	0.50	0.00	-0.235	-0.233	-0.252	-0.08477	-0.15555	0.51874
660	2	1.08	-0.00	-0.227	-0.229	-0.250	-0.08203	-0.15334	0.51532
660	3	0.06	-0.00	-0.225	-0.235	-0.244	-0.08109	-0.15334	0.50891
660	4	0.48	-0.00	-0.225	-0.237	-0.243	-0.08135	-0.15419	0.50915
660	5	1.08	-0.00	-0.237	-0.251	-0.254	-0.08560	-0.16189	0.52384
660	6	0.07	-0.00	-0.229	-0.237	-0.247	-0.08246	-0.15531	0.51309
660	7	0.17	-0.00	-0.280	-0.281	-0.301	-0.10099	-0.18664	0.53032
660	8	9.30	-0.00	-0.273	-0.268	-0.291	-0.09859	-0.17951	0.53814
660	9	12.52	-0.00	-0.298	-0.298	-0.333	-0.10733	-0.20248	0.56205
660	10	0.06	-0.00	-0.228	-0.235	-0.245	-0.08228	-0.15409	0.51534
661	1	0.51	0.00	-0.226	-0.229	-0.243	-0.08145	-0.14955	0.50727
661	2	1.07	-0.00	-0.232	-0.243	-0.249	-0.08126	-0.15192	0.51246
661	3	0.06	-0.00	-0.232	-0.243	-0.249	-0.08376	-0.15751	0.51735
661	4	0.48	-0.00	-0.234	-0.237	-0.241	-0.08038	-0.15322	0.50646
661	5	1.09	-0.00	-0.232	-0.247	-0.254	-0.08472	-0.15988	0.52319
661	10	0.50	-0.00	-0.246	-0.248	-0.270	-0.08886	-0.16491	0.52760
661	11	0.52	-0.00	-0.267	-0.261	-0.285	-0.09617	-0.17491	0.53760
661	12	0.54	-0.00	-0.309	-0.311	-0.344	-0.11356	-0.21021	0.58130
661	13	0.05	-0.00	-0.232	-0.244	-0.251	-0.08378	-0.15882	0.51752
662	1	0.48	0.00	-0.244	-0.244	-0.262	-0.08792	-0.16217	0.52449
662	2	1.05	-0.00	-0.220	-0.225	-0.241	-0.07920	-0.14939	0.50164
662	3	0.05	-0.00	-0.222	-0.233	-0.242	-0.08026	-0.15250	0.50882
662	4	0.47	-0.00	-0.229	-0.242	-0.247	-0.08256	-0.15679	0.51480

TABLE VA
BALANCE CAVITY AND BASE PRESSURE COEFFICIENTS, AND INCREMENTAL
AXIAL FORCE COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	M _c	α	φ	C _p c	C _p b ₁	C _p b ₂	ΔC _A c	ΔC _A b	C _A u
662	1.051	0.99	-0.00	-0.223	-0.237	-0.242	-0.08061	-0.15343	0.51243
662	1.050	3.10	-0.00	-0.237	-0.243	-0.257	-0.08550	-0.16022	0.52126
662	1.053	6.20	-0.00	-0.244	-0.244	-0.267	-0.08798	-0.16376	0.51728
662	1.051	9.34	-0.00	-0.275	-0.268	-0.292	-0.09921	-0.17959	0.55130
662	1.053	12.55	-0.00	-0.297	-0.302	-0.329	-0.10726	-0.20227	0.56536
662	1.053	-0.01	-0.00	-0.215	-0.227	-0.234	-0.07751	-0.14768	0.49697
663	1.052	3.09	0.00	-0.221	-0.221	-0.241	-0.07980	-0.14798	0.49964
663	1.050	-1.04	-0.00	-0.225	-0.231	-0.245	-0.08106	-0.15265	0.51153
663	1.051	-0.03	-0.00	-0.219	-0.230	-0.237	-0.07995	-0.14996	0.50818
663	1.051	0.47	-0.00	-0.222	-0.236	-0.240	-0.07994	-0.15247	0.51193
663	1.050	1.02	-0.00	-0.222	-0.230	-0.243	-0.08318	-0.15768	0.51691
663	1.049	3.15	-0.00	-0.221	-0.230	-0.245	-0.08001	-0.15238	0.50573
663	1.051	6.26	-0.00	-0.258	-0.256	-0.281	-0.09306	-0.17238	0.55437
663	1.051	9.35	-0.00	-0.278	-0.274	-0.297	-0.10039	-0.18291	0.55997
663	1.053	12.55	-0.00	-0.293	-0.299	-0.323	-0.10579	-0.19317	0.55452
663	1.051	-0.04	-0.00	-0.225	-0.237	-0.243	-0.08115	-0.15405	0.50562
664	1.050	3.08	0.00	-0.236	-0.237	-0.257	-0.08506	-0.15845	0.50766
664	1.049	-1.03	-0.00	-0.240	-0.245	-0.259	-0.08657	-0.16482	0.52149
664	1.049	0.52	-0.00	-0.238	-0.252	-0.256	-0.08580	-0.16267	0.52390
664	1.052	1.04	-0.00	-0.232	-0.246	-0.251	-0.08364	-0.15935	0.51332
664	1.050	3.19	-0.00	-0.251	-0.247	-0.270	-0.09046	-0.16733	0.53809
664	1.047	6.27	-0.00	-0.289	-0.286	-0.314	-0.10418	-0.19110	0.57428
664	1.052	9.36	-0.00	-0.297	-0.303	-0.324	-0.10721	-0.20553	0.57428
664	1.050	12.56	-0.00	-0.226	-0.240	-0.244	-0.08147	-0.15553	0.51785
665	1.050	3.09	0.00	-0.230	-0.234	-0.255	-0.08292	-0.15679	0.51437
665	1.053	-1.02	-0.00	-0.223	-0.238	-0.248	-0.08040	-0.15578	0.51487
665	1.051	0.52	-0.00	-0.246	-0.245	-0.246	-0.08166	-0.15698	0.52385
665	1.050	1.06	-0.00	-0.241	-0.257	-0.259	-0.08502	-0.16298	0.53009
665	1.049	3.16	-0.00	-0.237	-0.253	-0.259	-0.08914	-0.16362	0.53809
665	1.050	6.25	-0.00	-0.255	-0.248	-0.278	-0.09198	-0.16888	0.54713
665	1.050	9.35	-0.00	-0.275	-0.275	-0.297	-0.09934	-0.18328	0.57135
665	1.050	12.56	-0.00	-0.313	-0.320	-0.333	-0.12779	-0.20920	0.59073
665	1.050	-0.00	-0.00	-0.240	-0.257	-0.260	-0.08664	-0.16575	0.53323
666	1.050	3.05	-0.00	-0.232	-0.235	-0.254	-0.08369	-0.15704	0.52087
666	1.049	-1.02	-0.00	-0.239	-0.235	-0.265	-0.08654	-0.16647	0.54469

TABLE VA
BALANCE CAVITY AND BASE PRESSURE COEFFICIENTS, AND INCREMENTAL
AXIAL FORCE COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	M _c	α	Φ	C _{p,c}	C _{p,b1}	C _{p,b2}	ΔC _{A,c}	ΔC _{A,b}	C _{A,u}
666	3	1.050	0.02	-0.00	-0.237	-0.254	-0.257	-0.06553	-0.16385	0.54519
666	4	1.052	0.03	-0.00	-0.230	-0.249	-0.249	-0.06314	-0.15958	0.53766
666	5	1.051	1.09	-0.00	-0.245	-0.261	-0.264	-0.06635	-0.16933	0.54307
666	6	1.048	3.18	-0.00	-0.250	-0.256	-0.272	-0.09222	-0.16905	0.56862
666	7	1.050	6.31	-0.00	-0.256	-0.256	-0.279	-0.09247	-0.16960	0.57802
666	8	1.051	9.37	-0.00	-0.274	-0.274	-0.298	-0.09877	-0.18351	0.57802
666	9	1.050	12.58	-0.00	-0.311	-0.318	-0.333	-0.11196	-0.20861	0.57106
666	10	1.052	10.03	-0.00	-0.233	-0.251	-0.255	-0.06395	-0.16218	0.53334
667	1	0.52	-3.01	0.00	-0.230	-0.234	-0.253	-0.06297	-0.15600	0.52705
667	2	0.51	-0.97	-0.00	-0.237	-0.250	-0.264	-0.06542	-0.16475	0.54565
667	3	0.50	0.47	-0.00	-0.243	-0.256	-0.261	-0.06664	-0.16563	0.54736
667	4	0.49	0.57	-0.00	-0.237	-0.253	-0.262	-0.06764	-0.16603	0.54947
667	5	0.53	1.10	-0.00	-0.244	-0.255	-0.256	-0.06553	-0.16509	0.54991
667	6	0.52	3.21	-0.00	-0.271	-0.261	-0.268	-0.06799	-0.17703	0.54977
667	7	0.48	6.28	-0.00	-0.273	-0.263	-0.295	-0.09843	-0.18958	0.53040
667	8	0.52	9.39	-0.00	-0.273	-0.263	-0.295	-0.09843	-0.18958	0.53040
667	9	0.50	12.60	-0.00	-0.313	-0.273	-0.332	-0.12283	-0.20958	0.52228
667	10	0.50	10.06	-0.00	-0.243	-0.257	-0.262	-0.06749	-0.16628	0.54488
668	1	0.50	3.00	0.00	-0.241	-0.243	-0.261	-0.06704	-0.16169	0.54891
668	2	0.50	0.93	-0.00	-0.246	-0.259	-0.277	-0.06885	-0.17160	0.56871
668	3	0.49	0.09	-0.00	-0.247	-0.263	-0.269	-0.06923	-0.17067	0.56893
668	4	0.50	0.57	-0.00	-0.243	-0.258	-0.263	-0.06797	-0.16706	0.57479
668	5	0.47	1.11	-0.00	-0.251	-0.277	-0.272	-0.09473	-0.17739	0.59419
668	6	0.51	3.20	-0.00	-0.261	-0.251	-0.283	-0.09411	-0.17133	0.59403
668	7	0.50	6.31	-0.00	-0.264	-0.268	-0.308	-0.10236	-0.19102	0.59403
668	8	0.49	9.39	-0.00	-0.284	-0.288	-0.341	-0.11569	-0.21562	0.62408
668	9	0.49	12.60	-0.00	-0.321	-0.332	-0.341	-0.11569	-0.21562	0.62408
668	10	0.51	10.05	-0.00	-0.244	-0.260	-0.266	-0.06794	-0.16862	0.57130
669	5	0.48	-3.11	0.00	-0.241	-0.258	-0.261	-0.06880	-0.16000	0.53535
669	6	0.45	-0.12	-0.00	-0.245	-0.258	-0.263	-0.06836	-0.16690	0.53535
669	7	0.45	0.51	-0.00	-0.252	-0.265	-0.244	-0.06258	-0.15592	0.51924
669	8	0.43	-0.01	-0.00	-0.265	-0.265	-0.269	-0.09107	-0.17142	0.54908
669	9	0.43	1.07	-0.00	-0.270	-0.270	-0.260	-0.08370	-0.17242	0.54908
669	10	0.50	3.07	-0.00	-0.232	-0.245	-0.252	-0.08429	-0.15923	0.51798
669	11	0.50	3.11	-0.00	-0.234	-0.244	-0.252	-0.08429	-0.15923	0.51798
669	12	0.48	4.15	-0.00	-0.250	-0.260	-0.267	-0.09008	-0.16903	0.53074
669	13	0.49	-0.04	-0.00	-0.229	-0.239	-0.246	-0.08244	-0.15545	0.51449
670	3	1.047	-3.10	-0.00	-0.248	-0.246	-0.263	-0.06948	-0.16326	0.52489

TABLE VA
 BALANCE CAVITY AND BASE PRESSURE COEFFICIENTS, AND INCREMENTAL
 AXIAL FORCE COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	M _c	α	Φ	C _{p,c}	C _{p,b1}	C _{p,b2}	ΔC _{A,c}	ΔC _{A,b}	C _{A,u}
670	4	1.047	-0.04	-0.00	-0.238	-0.250	-0.257	-0.08595	-0.16254	0.52534
670	5	1.049	0.47	-0.00	-0.228	-0.241	-0.243	-0.08227	-0.15533	0.50938
670	6	1.049	1.00	-0.00	-0.240	-0.252	-0.252	-0.08650	-0.16231	0.52478
670	7	1.049	2.07	-0.00	-0.234	-0.247	-0.246	-0.08441	-0.15809	0.51884
670	8	1.048	3.12	-0.00	-0.237	-0.244	-0.249	-0.08557	-0.15835	0.52345
670	9	1.048	4.18	-0.00	-0.248	-0.252	-0.261	-0.08935	-0.16466	0.52377
670	10	1.045	-0.03	-0.00	-0.244	-0.255	-0.260	-0.08795	-0.16528	0.52777
671	1	1.051	-3.10	-0.00	-0.223	-0.225	-0.242	-0.08055	-0.14982	0.50216
671	2	1.050	-0.49	-0.00	-0.231	-0.240	-0.249	-0.08326	-0.15666	0.50526
671	3	1.047	0.99	-0.00	-0.226	-0.237	-0.243	-0.08142	-0.15380	0.50820
671	4	1.049	2.06	-0.00	-0.236	-0.246	-0.251	-0.08509	-0.15949	0.51712
671	5	1.048	3.12	-0.00	-0.238	-0.248	-0.256	-0.08595	-0.16147	0.51461
671	6	1.044	4.14	-0.00	-0.248	-0.253	-0.266	-0.08918	-0.16674	0.51483
671	7	1.049	-0.02	-0.00	-0.244	-0.247	-0.266	-0.08818	-0.16454	0.51034
671	8	1.049	-0.02	-0.00	-0.233	-0.242	-0.249	-0.08403	-0.15730	0.51034
672	1	1.048	-3.09	-0.00	-0.240	-0.241	-0.257	-0.08658	-0.15941	0.51538
672	2	1.045	-0.47	-0.00	-0.254	-0.262	-0.268	-0.09158	-0.16992	0.53692
672	3	1.049	1.03	-0.00	-0.226	-0.236	-0.241	-0.08159	-0.15306	0.50789
672	4	1.048	2.06	-0.00	-0.236	-0.249	-0.251	-0.08508	-0.16123	0.52104
672	5	1.048	3.15	-0.00	-0.241	-0.251	-0.254	-0.08681	-0.16113	0.52151
672	6	1.049	4.15	-0.00	-0.246	-0.254	-0.262	-0.08891	-0.16523	0.52166
672	7	1.049	-0.03	-0.00	-0.245	-0.244	-0.262	-0.08842	-0.16236	0.51667
672	8	1.048	-0.03	-0.00	-0.233	-0.242	-0.249	-0.08423	-0.15728	0.51667
673	1	1.046	-3.11	-0.00	-0.244	-0.244	-0.259	-0.08789	-0.16117	0.52143
673	2	1.044	-0.47	-0.00	-0.245	-0.252	-0.260	-0.08850	-0.16422	0.53193
673	3	1.049	1.02	-0.00	-0.231	-0.240	-0.246	-0.08337	-0.15588	0.50264
673	4	1.045	2.05	-0.00	-0.243	-0.252	-0.260	-0.08783	-0.16414	0.52240
673	5	1.045	3.09	-0.00	-0.243	-0.252	-0.261	-0.08753	-0.16377	0.52200
673	6	1.048	4.15	-0.00	-0.248	-0.252	-0.269	-0.08952	-0.16707	0.52999
673	7	1.048	-0.03	-0.00	-0.240	-0.239	-0.261	-0.08654	-0.16057	0.51699
673	8	1.050	-0.03	-0.00	-0.224	-0.231	-0.241	-0.08084	-0.15106	0.49924
674	1	1.048	-3.09	-0.00	-0.229	-0.225	-0.245	-0.08269	-0.15193	0.50454
674	2	1.050	-0.47	-0.00	-0.228	-0.235	-0.243	-0.08217	-0.15331	0.50149
674	3	1.049	1.02	-0.00	-0.234	-0.242	-0.247	-0.08440	-0.15669	0.50292
674	4	1.048	2.05	-0.00	-0.239	-0.246	-0.244	-0.08255	-0.15424	0.51574
674	5	1.048	3.15	-0.00	-0.232	-0.246	-0.255	-0.08588	-0.16075	0.51737
674	6	1.049	4.15	-0.00	-0.237	-0.256	-0.271	-0.09097	-0.16875	0.52734
674	7	1.049	-0.02	-0.00	-0.233	-0.239	-0.257	-0.08546	-0.15795	0.51254
674	8	1.047	-0.02	-0.00	-0.233	-0.239	-0.249	-0.08429	-0.15645	0.51170

TABLE VA
 BALANCE CAVITY AND BASE PRESSURE COEFFICIENTS, AND INCREMENTAL
 AXIAL FORCE COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	M _c	α	φ	C _{p_c}	C _{p_{b1}}	C _{p_{b2}}	ΔC _{A_c}	ΔC _{A_b}	C _{A_u}
675	1.051	-3.09	0.00	-0.233	-0.242	-0.251	-0.08404	-0.15789	0.51049
675	1.050	-0.03	-0.00	-0.232	-0.243	-0.251	-0.08374	-0.15862	0.51608
675	1.053	0.47	-0.00	-0.223	-0.236	-0.243	-0.08056	-0.15362	0.50856
675	1.051	1.03	-0.00	-0.228	-0.242	-0.251	-0.08221	-0.15807	0.51030
675	1.050	2.06	-0.00	-0.232	-0.244	-0.259	-0.08367	-0.16138	0.51982
675	1.052	3.14	-0.00	-0.229	-0.239	-0.258	-0.08276	-0.15928	0.51366
675	1.053	4.16	-0.00	-0.229	-0.234	-0.257	-0.08251	-0.15745	0.51023
675	1.054	-0.01	-0.00	-0.222	-0.233	-0.240	-0.07999	-0.15157	0.50093
676	1.052	-3.11	-0.00	-0.228	-0.234	-0.242	-0.08213	-0.15252	0.50543
676	1.051	-0.01	-0.00	-0.221	-0.234	-0.239	-0.07960	-0.15164	0.50689
676	1.052	0.48	-0.00	-0.225	-0.239	-0.244	-0.08133	-0.15504	0.50850
676	1.053	1.07	-0.00	-0.232	-0.239	-0.255	-0.08371	-0.15988	0.50438
676	1.051	3.13	-0.00	-0.232	-0.244	-0.258	-0.08356	-0.15988	0.51164
676	1.053	4.16	-0.00	-0.233	-0.240	-0.258	-0.08387	-0.16100	0.51421
676	1.050	-0.03	-0.00	-0.228	-0.241	-0.247	-0.08423	-0.15970	0.51200
677	1.054	-3.10	-0.00	-0.224	-0.238	-0.241	-0.08096	-0.15359	0.51181
677	1.054	-0.03	-0.00	-0.222	-0.234	-0.247	-0.08010	-0.15448	0.50754
677	1.049	0.48	-0.00	-0.235	-0.248	-0.261	-0.08493	-0.16306	0.52961
677	1.051	1.01	-0.00	-0.234	-0.257	-0.267	-0.08533	-0.16568	0.53091
677	1.051	2.06	-0.00	-0.234	-0.242	-0.275	-0.08450	-0.16747	0.52041
677	1.050	3.12	-0.00	-0.230	-0.242	-0.267	-0.08306	-0.16323	0.51997
677	1.049	4.15	-0.00	-0.231	-0.241	-0.259	-0.08350	-0.16047	0.51061
677	1.049	-0.02	-0.00	-0.234	-0.247	-0.258	-0.08441	-0.16198	0.52280
67A	1.051	-3.10	-0.00	-0.236	-0.242	-0.253	-0.08502	-0.15852	0.52290
67A	1.053	-0.02	-0.00	-0.231	-0.244	-0.252	-0.08527	-0.15911	0.52028
67A	1.047	0.50	-0.00	-0.238	-0.251	-0.260	-0.08588	-0.16393	0.52815
67A	1.050	1.09	-0.00	-0.243	-0.260	-0.279	-0.09273	-0.17066	0.53107
67A	1.047	3.12	-0.00	-0.242	-0.260	-0.273	-0.08765	-0.16728	0.53468
67A	1.050	4.16	-0.00	-0.238	-0.259	-0.263	-0.08735	-0.16728	0.53468
67B	1.049	-0.01	-0.00	-0.246	-0.258	-0.264	-0.08874	-0.16752	0.53595
679	1.051	-3.15	0.00	-0.241	-0.247	-0.260	-0.08710	-0.16151	0.53302
679	1.052	-0.07	-0.00	-0.234	-0.247	-0.256	-0.08428	-0.16103	0.53370
679	1.050	0.98	-0.00	-0.229	-0.243	-0.250	-0.08269	-0.15792	0.52189
679	1.048	3.03	-0.00	-0.233	-0.248	-0.251	-0.08393	-0.15336	0.52189
679	1.047	3.09	-0.00	-0.238	-0.254	-0.255	-0.08598	-0.16338	0.52658
679	1.047	3.09	-0.00	-0.240	-0.253	-0.256	-0.08668	-0.16338	0.52912

TABLE VA
BALANCE CAVITY AND BASE PRESSURE COEFFICIENTS, AND INCREMENTAL
AXIAL FORCE COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	M _c	α	Φ	C _{Pc}	C _{Pb1}	C _{Pb2}	ΔC _{Ac}	ΔC _{Ab}	C _{Au}
679 9	1.047	4.12	-0.00	-0.248	-0.257	-0.266	-0.08963	-0.16700	0.53077
679 10	1.049	-0.05	-0.00	-0.236	-0.247	-0.256	-0.08506	-0.16132	0.52557
680 1	1.050	-3.11	-0.00	-0.237	-0.241	-0.256	-0.08537	-0.15959	0.52117
680 2	1.050	-0.03	-0.00	-0.235	-0.247	-0.254	-0.08494	-0.16040	0.51904
680 3	1.048	0.02	-0.00	-0.236	-0.248	-0.254	-0.08514	-0.16005	0.52022
680 4	1.048	1.00	-0.00	-0.237	-0.250	-0.253	-0.08544	-0.16132	0.52453
680 5	1.050	2.05	-0.00	-0.242	-0.253	-0.256	-0.08717	-0.16336	0.52400
680 6	1.049	3.09	-0.00	-0.243	-0.251	-0.257	-0.08756	-0.16285	0.52424
680 7	1.045	4.15	-0.00	-0.246	-0.243	-0.275	-0.09247	-0.17253	0.52760
680 8	1.050	-2.12	-0.00	-0.241	-0.243	-0.260	-0.08697	-0.16144	0.52542
680 9	1.048	-0.03	-0.00	-0.237	-0.248	-0.254	-0.08550	-0.16100	0.52451
681 1	1.047	-3.12	-0.00	-0.243	-0.249	-0.260	-0.08777	-0.16140	0.52559
681 2	1.048	-0.02	-0.00	-0.237	-0.249	-0.254	-0.08532	-0.16141	0.52371
681 3	1.048	0.02	-0.00	-0.239	-0.250	-0.255	-0.08609	-0.16175	0.52363
681 4	1.048	1.01	-0.00	-0.240	-0.254	-0.255	-0.08649	-0.16306	0.52631
681 5	1.050	2.05	-0.00	-0.237	-0.248	-0.251	-0.08541	-0.16007	0.52244
681 6	1.047	3.11	-0.00	-0.244	-0.251	-0.260	-0.08795	-0.16397	0.52972
681 7	1.047	4.14	-0.00	-0.251	-0.254	-0.260	-0.09041	-0.16730	0.53335
681 8	1.050	-0.03	-0.00	-0.234	-0.245	-0.251	-0.08453	-0.15807	0.51599
682 1	1.050	-3.10	-0.00	-0.234	-0.236	-0.252	-0.08448	-0.15619	0.51543
682 2	1.048	-0.01	-0.00	-0.237	-0.248	-0.252	-0.08533	-0.16020	0.52402
682 3	1.048	1.03	-0.00	-0.243	-0.255	-0.259	-0.08782	-0.16491	0.52685
682 4	1.050	2.08	-0.00	-0.236	-0.249	-0.250	-0.08525	-0.16000	0.52335
682 5	1.049	3.14	-0.00	-0.243	-0.253	-0.257	-0.08763	-0.16380	0.53001
682 6	1.050	4.17	-0.00	-0.237	-0.243	-0.252	-0.08534	-0.15861	0.52602
682 7	1.048	-0.02	-0.00	-0.249	-0.251	-0.267	-0.08964	-0.16614	0.53596
682 8	1.047	-0.03	-0.00	-0.240	-0.250	-0.255	-0.08660	-0.16109	0.52709
683 10	0.898	-3.11	-0.00	-0.122	-0.124	-0.139	-0.04421	-0.08314	0.30011
683 11	0.897	-1.07	-0.00	-0.120	-0.124	-0.144	-0.04322	-0.08594	0.29299
683 13	0.898	-0.47	-0.00	-0.118	-0.130	-0.139	-0.04274	-0.08653	0.29177
683 14	0.897	0.96	-0.00	-0.121	-0.133	-0.141	-0.04381	-0.08809	0.29101
683 15	0.898	3.07	-0.00	-0.122	-0.134	-0.140	-0.04417	-0.08803	0.29101
683 16	0.897	6.17	-0.00	-0.130	-0.134	-0.163	-0.04715	-0.09541	0.29161
683 17	0.897	9.27	-0.00	-0.152	-0.147	-0.196	-0.05506	-0.10980	0.30462
683 18	0.897	12.46	-0.00	-0.190	-0.179	-0.232	-0.06042	-0.13177	0.33452
683 19	0.899	-0.06	-0.00	-0.113	-0.126	-0.136	-0.04085	-0.08415	0.29222

TABLE VA
BALANCE CAVITY AND BASE PRESSURE COEFFICIENTS, AND INCREMENTAL
AXIAL FORCE COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	M _c	α	φ	C _p c	C _p b ₁	C _p b ₂	ΔC _A c	ΔC _A b	C _A u
684	0.896	-3.10	-0.00	-0.124	-0.124	-0.141	-0.04466	-0.08529	0.29208
684	0.899	-1.08	-0.00	-0.114	-0.120	-0.138	-0.04124	-0.08303	0.29155
684	0.897	-0.05	-0.00	-0.115	-0.128	-0.137	-0.04158	-0.08493	0.28937
684	0.899	0.50	-0.00	-0.113	-0.130	-0.138	-0.04096	-0.08437	0.28765
684	0.896	0.99	-0.00	-0.121	-0.137	-0.135	-0.04365	-0.08821	0.28894
684	0.901	3.09	-0.00	-0.119	-0.131	-0.135	-0.04295	-0.08545	0.29529
684	0.89A	6.18	-0.00	-0.130	-0.134	-0.164	-0.04682	-0.09561	0.30583
684	0.89A	9.2A	-0.00	-0.110	-0.145	-0.193	-0.05409	-0.10869	0.34130
684	0.900	12.46	-0.00	-0.187	-0.179	-0.231	-0.06762	-0.13157	0.30583
684	0.900	-0.04	-0.00	-0.112	-0.124	-0.132	-0.04048	-0.08242	0.28839
685	0.900	3.10	-0.00	-0.117	-0.116	-0.134	-0.04225	-0.08039	0.29094
685	0.89A	-1.07	-0.00	-0.116	-0.129	-0.140	-0.04176	-0.08482	0.28669
685	0.89A	-0.03	-0.00	-0.116	-0.129	-0.136	-0.04200	-0.08528	0.28815
685	0.900	0.52	-0.00	-0.113	-0.130	-0.134	-0.04185	-0.08468	0.28535
685	0.897	1.09	-0.00	-0.120	-0.131	-0.141	-0.04092	-0.08370	0.29088
685	0.897	6.18	-0.00	-0.130	-0.133	-0.163	-0.04709	-0.09519	0.30001
685	0.898	9.30	-0.00	-0.148	-0.143	-0.192	-0.05333	-0.10721	0.30712
685	0.896	12.45	-0.00	-0.193	-0.183	-0.236	-0.06964	-0.13437	0.34120
685	0.897	-0.04	-0.00	-0.118	-0.133	-0.141	-0.04283	-0.08798	0.28870
686	0.897	3.06	-0.00	-0.118	-0.119	-0.138	-0.04265	-0.08282	0.28781
686	0.899	-1.04	-0.00	-0.115	-0.129	-0.137	-0.04391	-0.08251	0.28916
686	0.89A	-0.52	-0.00	-0.114	-0.128	-0.131	-0.04142	-0.08524	0.28734
686	0.896	1.03	-0.00	-0.118	-0.133	-0.136	-0.04267	-0.08660	0.28998
686	0.899	3.09	-0.00	-0.128	-0.132	-0.145	-0.04635	-0.08909	0.29930
686	0.89A	6.20	-0.00	-0.148	-0.147	-0.190	-0.05341	-0.10604	0.30503
686	0.899	9.30	-0.00	-0.168	-0.177	-0.229	-0.06786	-0.13041	0.30989
686	0.900	12.46	-0.00	-0.168	-0.177	-0.229	-0.06786	-0.13041	0.34348
686	0.89A	-0.03	-0.00	-0.114	-0.128	-0.139	-0.04130	-0.08429	0.28918
686	0.897	-3.08	-0.00	-0.114	-0.128	-0.141	-0.04135	-0.08286	0.28507
686	0.897	-1.03	-0.00	-0.114	-0.126	-0.141	-0.04135	-0.08590	0.29170
687	0.899	0.00	-0.00	-0.116	-0.132	-0.136	-0.04203	-0.08610	0.29176
687	0.899	0.52	-0.00	-0.116	-0.132	-0.135	-0.04281	-0.08629	0.29176
687	0.897	1.04	-0.00	-0.129	-0.132	-0.134	-0.04192	-0.08555	0.29519
687	0.897	3.11	-0.00	-0.129	-0.132	-0.147	-0.04574	-0.08970	0.30682
687	0.901	6.19	-0.00	-0.142	-0.135	-0.163	-0.05156	-0.09420	0.31151
687	0.899	9.31	-0.00	-0.168	-0.179	-0.231	-0.06776	-0.10313	0.35101
687	0.899	12.46	-0.00	-0.168	-0.179	-0.231	-0.06776	-0.10313	0.35101

TABLE VA
BALANCE CAVITY AND BASE PRESSURE COEFFICIENTS, AND INCREMENTAL AXIAL FORCE COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	M _c	α	φ	C _{Pc}	C _{Pb1}	C _{Pb2}	ΔC _{Ac}	ΔC _{Aδ}	C _{Au}
687	8	0.899	-0.01	-0.00	-0.115	-0.132	-0.135	-0.04170	-0.06574	0.29126
688A	1	0.89A	3.07	-0.00	-0.114	-0.120	-0.139	-0.04138	-0.06346	0.28730
688A	2	0.899	-1.00	-0.00	-0.110	-0.127	-0.138	-0.03964	-0.06515	0.29867
688A	3	0.899	-0.01	-0.00	-0.113	-0.135	-0.135	-0.04089	-0.06691	0.30214
688A	4	0.899	0.56	-0.00	-0.118	-0.138	-0.136	-0.04272	-0.06808	0.30358
688A	5	0.899	1.02	-0.00	-0.117	-0.138	-0.134	-0.04233	-0.06755	0.30355
688A	6	0.899	3.12	-0.00	-0.124	-0.130	-0.148	-0.04464	-0.06931	0.32013
688A	7	0.899	6.20	-0.00	-0.127	-0.136	-0.161	-0.04599	-0.09270	0.32147
688A	8	0.89A	9.30	-0.00	-0.143	-0.136	-0.186	-0.05163	-0.10346	0.35969
688A	9	0.89A	12.46	-0.00	-0.188	-0.177	-0.230	-0.06789	-0.13057	0.35069
688A	10	0.89A	-0.00	-0.00	-0.114	-0.137	-0.138	-0.04118	-0.06826	0.35069
689	1	0.897	3.02	-0.00	-0.116	-0.137	-0.146	-0.04211	-0.06627	0.29496
689	2	0.899	-0.99	-0.00	-0.114	-0.137	-0.149	-0.04120	-0.09176	0.31189
689	3	0.897	0.58	-0.00	-0.118	-0.145	-0.145	-0.04274	-0.09336	0.31533
689	4	0.901	0.57	-0.00	-0.117	-0.145	-0.138	-0.04216	-0.09038	0.31764
689	5	0.899	1.07	-0.00	-0.116	-0.139	-0.134	-0.04194	-0.06786	0.31804
689	6	0.899	3.14	-0.00	-0.123	-0.135	-0.147	-0.04462	-0.09476	0.33045
689	7	0.897	6.24	-0.00	-0.130	-0.132	-0.163	-0.04706	-0.09476	0.33336
689	8	0.897	9.33	-0.00	-0.144	-0.137	-0.189	-0.05187	-0.10435	0.33336
689	9	0.899	12.48	-0.00	-0.181	-0.173	-0.227	-0.06527	-0.12828	0.37042
689	10	0.897	-0.02	-0.00	-0.118	-0.147	-0.148	-0.04257	-0.09458	0.31424
690	3	0.899	3.00	-0.00	-0.117	-0.125	-0.147	-0.04223	-0.06721	0.30915
690	4	0.900	-0.98	-0.00	-0.117	-0.141	-0.154	-0.04229	-0.09474	0.32817
690	5	0.899	0.55	-0.00	-0.118	-0.148	-0.148	-0.04272	-0.09522	0.32990
690	6	0.899	1.06	-0.00	-0.118	-0.146	-0.143	-0.04266	-0.09109	0.33203
690	7	0.899	3.13	-0.00	-0.129	-0.145	-0.139	-0.04368	-0.09381	0.34172
690	8	0.89A	6.23	-0.00	-0.135	-0.140	-0.152	-0.04661	-0.09484	0.34542
690	9	0.89A	9.30	-0.00	-0.146	-0.139	-0.169	-0.05275	-0.10500	0.37722
690	10	0.89A	12.45	-0.00	-0.146	-0.139	-0.189	-0.05275	-0.10500	0.37722
690	11	0.89A	-0.03	-0.00	-0.183	-0.173	-0.230	-0.06658	-0.12930	0.32635
690	12	0.896	-0.03	-0.00	-0.123	-0.150	-0.153	-0.04451	-0.09709	0.32635
691	1	0.898	2.99	-0.00	-0.118	-0.126	-0.151	-0.04277	-0.06872	0.32124
691	2	0.897	-0.94	-0.00	-0.120	-0.149	-0.162	-0.04337	-0.09849	0.35647
691	3	0.898	0.56	-0.00	-0.118	-0.149	-0.155	-0.04347	-0.09764	0.34347
691	4	0.899	1.05	-0.00	-0.118	-0.148	-0.145	-0.04266	-0.09394	0.34347
691	5	0.899	3.15	-0.00	-0.121	-0.148	-0.141	-0.04356	-0.09306	0.34520
691	6	0.897	6.23	-0.00	-0.135	-0.145	-0.157	-0.04703	-0.09676	0.35201
691	7	0.897	-0.03	-0.00	-0.135	-0.135	-0.168	-0.04865	-0.09709	0.35756

TABLE VA
 BALANCE CAVITY AND BASE PRESSURE COEFFICIENTS, AND INCREMENTAL
 AXIAL FORCE COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	M _c	α	φ	C _p c	C _p b _l	C _p b ₂	ΔC _A c	ΔC _A b	C _A u
691	8	0.899	9.32	-0.00	-0.146	-0.140	-0.139	-0.04256	-0.08330	0.29435
691	9	0.89A	12.49	0.00	-0.187	-0.176	-0.137	-0.04066	-0.08592	0.29076
691	10	0.896	0.04	-0.00	-0.119	-0.149	-0.133	-0.04059	-0.08475	0.29099
692	1	0.899	-3.07	-0.00	-0.118	-0.121	-0.139	-0.04256	-0.08330	0.29435
692	2	0.89A	-0.03	-0.00	-0.113	-0.130	-0.137	-0.04066	-0.08592	0.29076
692	3	0.899	0.51	-0.00	-0.112	-0.131	-0.133	-0.04059	-0.08475	0.29099
692	4	0.899	1.02	-0.00	-0.115	-0.134	-0.138	-0.04151	-0.08728	0.29320
692	5	0.899	2.06	-0.00	-0.119	-0.136	-0.143	-0.04310	-0.08959	0.29847
692	6	0.899	3.09	-0.00	-0.121	-0.135	-0.149	-0.04373	-0.09103	0.29928
692	7	0.899	4.10	-0.00	-0.124	-0.137	-0.154	-0.04483	-0.09318	0.29986
692	8	0.900	-0.03	-0.00	-0.111	-0.128	-0.133	-0.04022	-0.08389	0.29174
693	1	0.896	-3.10	-0.00	-0.122	-0.124	-0.138	-0.04392	-0.08411	0.29282
693	2	0.900	-0.03	-0.00	-0.111	-0.127	-0.132	-0.04023	-0.08302	0.29026
693	3	0.900	0.51	-0.00	-0.113	-0.129	-0.130	-0.04029	-0.08331	0.28896
693	4	0.902	1.01	-0.00	-0.113	-0.132	-0.129	-0.04099	-0.08388	0.29703
693	5	0.899	2.06	-0.00	-0.116	-0.131	-0.129	-0.04177	-0.08351	0.30034
693	6	0.899	3.08	-0.00	-0.125	-0.138	-0.142	-0.04535	-0.08991	0.29838
693	7	0.89A	4.15	-0.00	-0.124	-0.133	-0.144	-0.04481	-0.08906	0.29838
693	8	0.897	-0.03	-0.00	-0.110	-0.128	-0.133	-0.03982	-0.08387	0.28901
694	1	0.89A	-3.09	-0.00	-0.117	-0.120	-0.138	-0.04242	-0.08296	0.28875
694	2	0.899	-0.03	-0.00	-0.108	-0.126	-0.131	-0.03923	-0.08259	0.28685
694	3	0.897	0.49	-0.00	-0.117	-0.133	-0.137	-0.04068	-0.08693	0.28897
694	4	0.89A	1.01	-0.00	-0.113	-0.131	-0.133	-0.04068	-0.08467	0.28868
694	5	0.899	2.04	-0.00	-0.116	-0.131	-0.137	-0.04193	-0.08601	0.29308
694	6	0.897	3.07	-0.00	-0.119	-0.131	-0.143	-0.04314	-0.08814	0.29433
694	7	0.897	4.10	-0.00	-0.122	-0.129	-0.148	-0.04420	-0.08895	0.29631
694	8	0.898	-0.03	-0.00	-0.108	-0.125	-0.132	-0.03904	-0.08258	0.28711
695	1	0.897	-3.10	-0.00	-0.116	-0.119	-0.136	-0.04182	-0.08184	0.28859
695	2	0.89A	-0.03	-0.00	-0.111	-0.129	-0.131	-0.03973	-0.08344	0.28745
695	3	0.897	0.50	-0.00	-0.110	-0.129	-0.131	-0.04005	-0.08368	0.28919
695	4	0.89A	1.02	-0.00	-0.112	-0.130	-0.130	-0.04061	-0.08368	0.29227
695	5	0.897	2.02	-0.00	-0.117	-0.134	-0.137	-0.04240	-0.08699	0.29664
695	6	0.89A	3.09	-0.00	-0.119	-0.129	-0.138	-0.04287	-0.08588	0.29703
695	7	0.898	4.13	-0.00	-0.121	-0.120	-0.146	-0.04367	-0.08842	0.28869
695	8	0.89A	-0.02	-0.00	-0.111	-0.126	-0.133	-0.04005	-0.08304	0.28699
696	1	0.899	-3.11	-0.00	-0.116	-0.121	-0.137	-0.04204	-0.08284	0.28939
696	2	0.897	-0.02	-0.00	-0.117	-0.131	-0.138	-0.04242	-0.08657	0.28718

TABLE VA
BALANCE CAVITY AND BASE PRESSURE COEFFICIENTS, AND INCREMENTAL
AXIAL FORCE COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	M _c	α	φ	C _p c	C _p b ₁	C _p b ₂	ΔC _A c	ΔC _A b	C _A u
696	0.898	0.51	-0.00	-0.111	-0.129	-0.134	-0.04026	-0.06432	0.28534
696	0.900	1.04	-0.00	-0.112	-0.130	-0.133	-0.04062	-0.06459	0.29008
696	0.900	2.04	-0.00	-0.117	-0.127	-0.143	-0.04298	-0.06827	0.29468
696	0.897	3.10	-0.00	-0.123	-0.133	-0.151	-0.04457	-0.09121	0.29644
696	0.89A	4.14	-0.00	-0.113	-0.127	-0.134	-0.04082	-0.06412	0.28645
697	0.897	-3.11	-0.00	-0.117	-0.119	-0.139	-0.04216	-0.06283	0.28927
697	0.89A	-0.02	-0.00	-0.114	-0.128	-0.136	-0.04126	-0.06496	0.28822
697	0.89A	0.50	-0.00	-0.111	-0.130	-0.131	-0.03999	-0.06377	0.28604
697	0.897	1.02	-0.00	-0.113	-0.131	-0.133	-0.04078	-0.064A3	0.28876
697	0.897	2.02	-0.00	-0.120	-0.135	-0.141	-0.04321	-0.06847	0.29306
697	0.89A	3.09	-0.00	-0.118	-0.130	-0.141	-0.04283	-0.06688	0.29618
697	0.899	4.14	-0.00	-0.119	-0.127	-0.143	-0.04289	-0.06678	0.29705
697	0.897	-0.03	-0.00	-0.111	-0.126	-0.134	-0.04022	-0.06360	0.28654
69A	0.89A	3.07	-0.00	-0.119	-0.129	-0.141	-0.04319	-0.06679	0.29176
69A	0.899	-0.02	-0.00	-0.110	-0.127	-0.133	-0.03965	-0.06343	0.28910
69A	0.899	0.49	-0.00	-0.111	-0.130	-0.135	-0.04023	-0.06482	0.28864
69A	0.895	1.01	-0.00	-0.112	-0.130	-0.137	-0.04038	-0.06563	0.28821
69A	0.899	2.05	-0.00	-0.116	-0.131	-0.144	-0.04183	-0.06853	0.29470
69A	0.89A	3.13	-0.00	-0.121	-0.131	-0.151	-0.04371	-0.09047	0.29715
69A	0.89A	4.12	-0.00	-0.122	-0.131	-0.153	-0.04408	-0.09131	0.29926
69A	0.89A	-0.02	-0.00	-0.111	-0.129	-0.137	-0.04008	-0.06521	0.28699
699	0.898	3.07	-0.00	-0.116	-0.126	-0.137	-0.04180	-0.06280	0.29049
699	0.900	-0.01	-0.00	-0.111	-0.126	-0.132	-0.03997	-0.06290	0.28895
699	0.899	0.49	-0.00	-0.112	-0.131	-0.134	-0.04033	-0.06483	0.28991
699	0.900	1.01	-0.00	-0.114	-0.134	-0.135	-0.04127	-0.06617	0.29215
699	0.89A	2.06	-0.00	-0.121	-0.138	-0.144	-0.04391	-0.09072	0.29520
699	0.899	3.12	-0.00	-0.119	-0.132	-0.144	-0.04307	-0.06902	0.29797
699	0.899	4.12	-0.00	-0.122	-0.132	-0.148	-0.04409	-0.06992	0.30057
699	0.899	-2.12	-0.00	-0.113	-0.120	-0.142	-0.04102	-0.06404	0.29115
699	0.900	-0.02	-0.00	-0.110	-0.126	-0.130	-0.03972	-0.06230	0.28862
700	0.899	3.09	-0.00	-0.119	-0.136	-0.137	-0.04307	-0.06778	0.30189
700	0.899	-0.06	-0.00	-0.110	-0.128	-0.135	-0.03976	-0.06427	0.29445
700	0.901	0.58	-0.00	-0.111	-0.131	-0.141	-0.04026	-0.06746	0.29799
700	0.899	2.06	-0.00	-0.116	-0.138	-0.152	-0.04182	-0.09217	0.30360
700	0.900	3.11	-0.00	-0.120	-0.133	-0.157	-0.04333	-0.09324	0.30446
700	0.89A	4.14	-0.00	-0.125	-0.138	-0.161	-0.04521	-0.09609	0.30392

TABLE VA
BALANCE CAVITY AND BASE PRESSURE COEFFICIENTS, AND INCREMENTAL
AXIAL FORCE COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	M _c	α	Φ	C _{Pc}	C _{Pb1}	C _{Pb2}	ΔC _{Ac}	ΔC _{Ab}	C _{Au}
700	0.901	-0.01	-0.00	-0.113	-0.129	-0.136	-0.04079	-0.08519	0.29522
701	0.899	-3.10	-0.00	-0.125	-0.126	-0.144	-0.04501	-0.08654	0.30555
701	0.900	-0.04	-0.00	-0.114	-0.133	-0.135	-0.04121	-0.08630	0.30292
701	0.898	0.52	-0.00	-0.117	-0.137	-0.139	-0.04219	-0.08860	0.30197
701	0.896	1.01	-0.00	-0.122	-0.141	-0.148	-0.04395	-0.09296	0.30402
701	0.897	2.05	-0.00	-0.125	-0.142	-0.149	-0.04527	-0.09358	0.30638
701	0.896	3.08	-0.00	-0.125	-0.143	-0.143	-0.04520	-0.09170	0.30910
701	0.896	4.14	-0.00	-0.124	-0.138	-0.143	-0.04497	-0.09036	0.30910
701	0.899	-0.03	-0.00	-0.113	-0.132	-0.136	-0.04091	-0.08590	0.30150
702	0.899	-3.16	44.99	-0.119	-0.129	-0.146	-0.04307	-0.08806	0.30385
702	0.897	-0.07	44.99	-0.119	-0.129	-0.138	-0.04302	-0.08581	0.29163
702	0.899	0.95	45.00	-0.115	-0.130	-0.137	-0.04143	-0.08579	0.29396
702	0.898	3.03	44.99	-0.118	-0.142	-0.149	-0.04448	-0.09368	0.29253
702	0.898	6.14	44.99	-0.127	-0.177	-0.188	-0.04576	-0.11594	0.29356
702	0.900	9.29	44.99	-0.141	-0.197	-0.212	-0.05102	-0.13131	0.28917
702	0.899	12.42	44.99	-0.158	-0.213	-0.196	-0.05704	-0.13100	0.27934
702	0.899	-0.06	44.99	-0.117	-0.126	-0.135	-0.04227	-0.08404	0.29100
703	0.897	-3.13	44.99	-0.117	-0.132	-0.145	-0.04242	-0.08903	0.28914
703	0.897	-0.01	44.99	-0.117	-0.131	-0.137	-0.04227	-0.08594	0.28703
703	0.899	0.98	44.99	-0.115	-0.135	-0.135	-0.04155	-0.08553	0.28867
703	0.900	3.07	44.99	-0.114	-0.135	-0.137	-0.04132	-0.08725	0.29258
703	0.900	6.18	44.99	-0.123	-0.165	-0.174	-0.04440	-0.10887	0.29782
703	0.899	9.30	44.99	-0.139	-0.194	-0.208	-0.05034	-0.12920	0.29594
703	0.897	12.43	44.99	-0.157	-0.211	-0.195	-0.05667	-0.13039	0.29011
703	0.898	-0.02	44.99	-0.113	-0.125	-0.136	-0.04092	-0.08380	0.28450
704	0.900	-3.12	44.99	-0.111	-0.128	-0.137	-0.04021	-0.08526	0.28746
704	0.898	-0.01	44.99	-0.113	-0.130	-0.133	-0.04093	-0.08448	0.28716
704	0.899	1.00	44.99	-0.116	-0.133	-0.135	-0.04178	-0.08597	0.28854
704	0.900	3.08	44.99	-0.121	-0.139	-0.139	-0.04368	-0.08907	0.29515
704	0.900	6.18	44.99	-0.127	-0.166	-0.178	-0.04582	-0.10336	0.29992
704	0.900	9.30	44.99	-0.134	-0.188	-0.202	-0.04826	-0.12533	0.29892
704	0.898	12.43	44.99	-0.158	-0.210	-0.195	-0.05722	-0.12998	0.29812
704	0.898	-0.01	44.99	-0.113	-0.127	-0.134	-0.04072	-0.08385	0.28339
705	0.899	-3.12	44.99	-0.113	-0.137	-0.142	-0.04085	-0.08957	0.28861
705	0.898	-0.01	44.99	-0.111	-0.127	-0.132	-0.04028	-0.08342	0.29220
705	0.897	1.01	44.99	-0.118	-0.136	-0.136	-0.04263	-0.08731	0.29428
705	0.897	3.12	44.99	-0.126	-0.143	-0.141	-0.04546	-0.09132	0.29660

TABLE VA
 BALANCE CAVITY AND BASE PRESSURE COEFFICIENTS, AND INCREMENTAL
 AXIAL FORCE COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	M _c	α	φ	C _p c	C _p b _l	C _p b ₂	ΔC _A c	ΔC _A b	C _A u
705	5	0.899	6.21	44.99	-0.123	-0.161	-0.171	-0.04433	-0.10676	0.31046
705	6	0.898	9.35	44.99	-0.138	-0.193	-0.205	-0.04991	-0.12757	0.30935
705	7	0.899	12.45	44.99	-0.153	-0.208	-0.189	-0.05530	-0.12749	0.30967
705	8	0.89A	0.00	44.99	-0.116	-0.132	-0.136	-0.04194	-0.08638	0.29205
706	1	0.900	-3.0A	44.99	-0.112	-0.143	-0.147	-0.04033	-0.09317	0.29914
706	2	0.900	1.02	44.99	-0.111	-0.129	-0.134	-0.03997	-0.08439	0.31362
706	3	0.899	3.14	44.99	-0.122	-0.142	-0.138	-0.04412	-0.08995	0.31647
706	4	0.89A	6.22	44.99	-0.133	-0.152	-0.149	-0.04788	-0.09635	0.32994
706	5	0.897	9.35	44.99	-0.129	-0.168	-0.173	-0.04672	-0.10943	0.32904
706	6	0.89A	12.47	44.99	-0.135	-0.167	-0.198	-0.04883	-0.12374	0.32844
706	7	0.899	0.03	44.99	-0.153	-0.206	-0.187	-0.05525	-0.12608	0.33007
706	8	0.89A	0.00	44.99	-0.116	-0.134	-0.137	-0.04185	-0.08688	0.33105
707	1	0.898	-3.04	44.99	-0.116	-0.149	-0.153	-0.04191	-0.09712	0.31445
707	2	0.900	1.08	44.99	-0.117	-0.135	-0.138	-0.04224	-0.08776	0.33808
707	3	0.899	3.15	44.99	-0.126	-0.148	-0.144	-0.04557	-0.09376	0.33857
707	4	0.900	6.24	44.99	-0.127	-0.155	-0.149	-0.04605	-0.09747	0.34813
707	5	0.899	9.36	44.99	-0.126	-0.163	-0.168	-0.04554	-0.10625	0.35279
707	6	0.899	12.48	44.99	-0.130	-0.182	-0.193	-0.04689	-0.12019	0.34931
707	7	0.899	0.04	44.99	-0.152	-0.204	-0.187	-0.05501	-0.12532	0.35171
707	8	0.899	0.00	44.99	-0.118	-0.136	-0.142	-0.04263	-0.08911	0.33362
70A	1	0.901	-2.99	44.99	-0.114	-0.141	-0.150	-0.04104	-0.09345	0.36495
70A	2	0.900	1.06	44.99	-0.119	-0.136	-0.145	-0.04291	-0.09027	0.39097
70A	3	0.900	3.17	44.99	-0.123	-0.145	-0.142	-0.04453	-0.09204	0.39977
70A	4	0.89A	6.27	44.99	-0.134	-0.159	-0.151	-0.04843	-0.09955	0.39783
70A	5	0.901	9.3A	44.99	-0.124	-0.152	-0.160	-0.04490	-0.10018	0.40383
70A	6	0.899	12.50	44.99	-0.132	-0.182	-0.199	-0.04778	-0.12228	0.40050
70A	7	0.900	0.08	44.99	-0.149	-0.203	-0.184	-0.05390	-0.12318	0.40149
70A	8	0.901	0.00	44.99	-0.115	-0.133	-0.140	-0.04166	-0.08765	0.39056
709	3	0.89A	3.13	90.00	-0.118	-0.141	-0.145	-0.04261	-0.09181	0.29085
709	4	0.899	-3.13	-78.80	-0.116	-0.138	-0.144	-0.04207	-0.09049	0.29121
709	5	0.900	-3.14	-56.50	-0.118	-0.140	-0.149	-0.04276	-0.09244	0.29045
709	6	0.901	-3.14	-45.00	-0.120	-0.139	-0.147	-0.04342	-0.09154	0.29038
709	7	0.89A	-3.14	-35.80	-0.115	-0.132	-0.141	-0.04153	-0.08751	0.29079
709	8	0.89A	-3.14	-22.50	-0.120	-0.125	-0.139	-0.04321	-0.08528	0.28948
709	9	0.89A	-3.14	-11.00	-0.119	-0.120	-0.138	-0.04316	-0.08285	0.28912
709	10	0.89A	-3.14	11.29	-0.118	-0.116	-0.135	-0.04352	-0.07996	0.28915
709	11	0.89A	-3.14	11.29	-0.121	-0.123	-0.135	-0.04386	-0.08289	0.29059

TABLE VA
 BALANCE CAVITY AND BASE PRESSURE COEFFICIENTS, AND INCREMENTAL
 AXIAL FORCE COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	M _c	α	Φ	C _{p,c}	C _{p,b1}	C _{p,b2}	ΔC _{A,c}	ΔC _{A,b}	C _{A,u}
709	13	0.900	-3.14	22.50	-0.117	-0.122	-0.135	-0.04230	-0.06239	0.29130
709	14	0.89A	-3.14	33.79	-0.117	-0.132	-0.141	-0.04378	-0.06770	0.29019
709	15	0.89A	-3.14	44.29	-0.117	-0.131	-0.142	-0.04239	-0.06767	0.29055
709	16	0.900	-3.14	56.29	-0.116	-0.131	-0.141	-0.04211	-0.06752	0.29157
709	17	0.899	-3.14	67.49	-0.115	-0.134	-0.142	-0.04162	-0.06641	0.29266
709	18	0.900	-3.14	78.79	-0.116	-0.139	-0.144	-0.04188	-0.09083	0.29249
709	19	0.899	-3.14	89.99	-0.121	-0.149	-0.151	-0.04362	-0.09624	0.29400
710	1	0.900	-0.04	89.99	-0.110	-0.123	-0.127	-0.03975	-0.06046	0.28716
710	2	0.899	-0.05	78.79	-0.117	-0.132	-0.135	-0.04218	-0.06571	0.28796
710	3	0.899	-0.05	67.49	-0.116	-0.130	-0.133	-0.04197	-0.06462	0.28831
710	4	0.896	-0.04	56.29	-0.113	-0.133	-0.138	-0.04386	-0.06692	0.28893
710	5	0.896	-0.05	44.29	-0.116	-0.127	-0.131	-0.04385	-0.06170	0.28733
710	6	0.898	-0.05	33.79	-0.116	-0.127	-0.135	-0.04178	-0.06422	0.28703
710	7	0.89A	-0.05	22.49	-0.116	-0.129	-0.135	-0.04203	-0.06405	0.28466
710	8	0.896	-0.05	11.29	-0.115	-0.129	-0.134	-0.04150	-0.06426	0.28646
710	9	0.899	-0.05	1.00	-0.115	-0.129	-0.133	-0.04166	-0.06420	0.28652
710	10	0.898	-0.05	-12.50	-0.119	-0.134	-0.136	-0.04292	-0.06681	0.28528
710	11	0.899	-0.05	-23.80	-0.115	-0.132	-0.133	-0.04144	-0.06509	0.28674
710	12	0.898	-0.05	-35.80	-0.112	-0.130	-0.131	-0.04165	-0.06379	0.28715
710	13	0.898	-0.06	-45.30	-0.116	-0.134	-0.136	-0.04204	-0.06662	0.28654
710	14	0.898	-0.05	-56.50	-0.116	-0.133	-0.135	-0.04190	-0.06601	0.28624
710	15	0.898	-0.05	-67.80	-0.118	-0.133	-0.137	-0.04278	-0.06742	0.28676
710	16	0.89A	-0.06	-78.80	-0.114	-0.129	-0.133	-0.04118	-0.06419	0.28660
710	17	0.898	-0.05	-90.00	-0.114	-0.129	-0.133	-0.04118	-0.06419	0.28660
711	1	0.898	3.07	90.00	-0.117	-0.146	-0.151	-0.04247	-0.09565	0.29564
711	2	0.899	3.06	-78.80	-0.118	-0.147	-0.148	-0.04255	-0.09569	0.29416
711	3	0.89A	3.05	-67.50	-0.122	-0.147	-0.146	-0.04205	-0.09593	0.29286
711	4	0.899	3.05	-56.30	-0.118	-0.141	-0.138	-0.04278	-0.08954	0.29372
711	5	0.897	3.05	-45.00	-0.119	-0.139	-0.134	-0.04315	-0.08784	0.29150
711	6	0.89A	3.05	-33.80	-0.122	-0.135	-0.134	-0.04304	-0.08630	0.29177
711	7	0.897	3.06	-22.50	-0.117	-0.131	-0.131	-0.04244	-0.08399	0.29239
711	8	0.89A	3.05	-11.29	-0.116	-0.127	-0.131	-0.04204	-0.08306	0.29497
711	9	0.897	3.06	1.00	-0.122	-0.131	-0.138	-0.04201	-0.08641	0.29393
711	10	0.897	3.06	11.29	-0.118	-0.132	-0.139	-0.04402	-0.08767	0.29301
711	11	0.89A	3.06	22.49	-0.119	-0.135	-0.139	-0.04248	-0.08647	0.29361
711	12	0.898	3.06	33.79	-0.119	-0.135	-0.141	-0.04296	-0.09074	0.29347
711	13	0.897	3.06	44.29	-0.122	-0.142	-0.141	-0.04395	-0.09074	0.29347
711	14	0.896	3.06	56.29	-0.116	-0.139	-0.136	-0.04346	-0.08638	0.29147
711	15	0.89A	3.06	67.49	-0.112	-0.143	-0.142	-0.04346	-0.09235	0.29473
711	16	0.899	3.05	78.79	-0.114	-0.146	-0.142	-0.04135	-0.09136	0.29661

TABLE VA
BALANCE CAVITY AND BASE PRESSURE COEFFICIENTS, AND INCREMENTAL
AXIAL FORCE COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	M _c	α	Φ	C _p c	C _p b ₁	C _p b ₂	ΔC _A c	ΔC _A b	C _A u
711	17	0.899	3.06	89.99	-0.119	-0.146	-0.149	-0.04288	-0.09480	0.29694
712	1	0.899	6.17	89.99	-0.127	-0.164	-0.183	-0.04588	-0.11440	0.30206
712	2	0.899A	6.16	78.79	-0.131	-0.167	-0.171	-0.04727	-0.10831	0.30016
712	3	0.899A	6.17	67.49	-0.130	-0.170	-0.159	-0.04696	-0.10573	0.30991
712	4	0.899A	6.16	56.29	-0.127	-0.170	-0.164	-0.04600	-0.10716	0.29990
712	5	0.899	6.15	44.99	-0.127	-0.168	-0.178	-0.04588	-0.11435	0.29891
712	6	0.896	6.15	33.79	-0.130	-0.163	-0.194	-0.04782	-0.11435	0.29616
712	7	0.899A	6.16	22.49	-0.131	-0.146	-0.193	-0.04703	-0.11058	0.29619
712	8	0.900	6.17	11.29	-0.123	-0.135	-0.175	-0.04662	-0.10823	0.30029
712	9	0.899A	6.17	-10.00	-0.126	-0.135	-0.163	-0.04565	-0.10956	0.30004
712	10	0.897	6.16	-12.50	-0.131	-0.127	-0.135	-0.04746	-0.10892	0.30906
712	11	0.899A	6.16	-23.80	-0.129	-0.156	-0.137	-0.04656	-0.10942	0.29934
712	12	0.897	6.16	-35.00	-0.123	-0.167	-0.138	-0.04653	-0.10978	0.29837
712	13	0.897	6.15	-45.30	-0.131	-0.187	-0.147	-0.04739	-0.10419	0.29678
712	14	0.897	6.16	-56.30	-0.129	-0.187	-0.159	-0.04673	-0.11119	0.29694
712	15	0.895	6.16	-67.80	-0.130	-0.182	-0.178	-0.04687	-0.11555	0.30044
712	16	0.896	6.16	-78.80	-0.128	-0.171	-0.178	-0.04616	-0.11188	0.29970
712	17	0.898	6.17	-90.00	-0.128	-0.171	-0.178	-0.04616	-0.11188	0.29970
713	1	0.897	9.31	90.00	-0.141	-0.192	-0.199	-0.05111	-0.12546	0.30393
713	2	0.899A	9.30	-78.79	-0.146	-0.210	-0.203	-0.05275	-0.13229	0.30235
713	3	0.899A	9.30	-78.79	-0.148	-0.210	-0.205	-0.05355	-0.13313	0.30268
713	4	0.899A	9.29	-67.49	-0.143	-0.208	-0.175	-0.05167	-0.12283	0.30929
713	5	0.900	9.28	-56.20	-0.137	-0.199	-0.151	-0.04957	-0.11126	0.29528
713	6	0.899	9.29	-45.00	-0.136	-0.189	-0.159	-0.04973	-0.11167	0.29508
713	7	0.899A	9.30	-33.80	-0.142	-0.174	-0.147	-0.04923	-0.10934	0.29877
713	8	0.899	9.30	-22.50	-0.149	-0.145	-0.147	-0.05112	-0.10944	0.30585
713	9	0.899A	9.31	-11.30	-0.142	-0.142	-0.154	-0.05393	-0.10613	0.30437
713	10	0.899A	9.29	0.00	-0.146	-0.154	-0.189	-0.05278	-0.11797	0.30378
713	11	0.897	9.30	11.30	-0.146	-0.142	-0.213	-0.05273	-0.11297	0.29519
713	12	0.899	9.29	22.49	-0.144	-0.154	-0.220	-0.05063	-0.12251	0.29519
713	13	0.896	9.28	33.79	-0.141	-0.168	-0.217	-0.05186	-0.13001	0.29565
713	14	0.896	9.29	44.99	-0.143	-0.198	-0.193	-0.05193	-0.12684	0.29507
713	15	0.896	9.28	56.29	-0.147	-0.203	-0.193	-0.05149	-0.12138	0.29507
713	16	0.897	9.29	67.49	-0.147	-0.198	-0.181	-0.05232	-0.12438	0.30658
713	17	0.899	9.29	78.79	-0.148	-0.191	-0.196	-0.05357	-0.13052	0.30589
713	18	0.897	9.29	89.99	-0.148	-0.191	-0.216	-0.05357	-0.13052	0.30589
714	1	0.897	12.41	90.00	-0.188	-0.261	-0.277	-0.06781	-0.17229	0.33920
714	2	0.896	12.42	78.78	-0.182	-0.236	-0.264	-0.06560	-0.16039	0.33623
714	3	0.896	12.41	67.48	-0.176	-0.227	-0.240	-0.06336	-0.14975	0.33183

TABLE VA
BALANCE CAVITY AND BASE PRESSURE COEFFICIENTS, AND INCREMENTAL
AXIAL FORCE COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	M _c	α	φ	C _{Pc}	C _{Pb1}	C _{Pb2}	ΔC _{Ac}	ΔC _{Ab}	C _{Au}	
714	4	0.897	12.41	56.29	-0.162	-0.205	-0.216	-0.05842	0.13500	0.29473
714	5	0.899	12.42	43.80	-0.158	-0.211	-0.193	-0.05667	0.12964	0.26955
714	6	0.897	12.39	22.51	-0.167	-0.194	-0.213	-0.05866	0.13059	0.29490
714	7	0.898	12.42	11.31	-0.177	-0.187	-0.249	-0.06373	0.13966	0.34011
714	8	0.899	12.42	11.00	-0.189	-0.178	-0.269	-0.06767	0.14342	0.34063
714	9	0.899	12.42	-11.31	-0.185	-0.180	-0.229	-0.06837	0.13107	0.33743
714	10	0.899	12.42	-22.51	-0.174	-0.190	-0.179	-0.06665	0.11830	0.32211
714	11	0.900	12.42	-33.80	-0.154	-0.150	-0.163	-0.06279	0.11032	0.29733
714	12	0.898	12.41	-45.29	-0.159	-0.201	-0.179	-0.05747	0.12203	0.29149
714	13	0.897	12.42	-56.49	-0.156	-0.218	-0.228	-0.05651	0.14284	0.29309
714	14	0.899	12.42	-67.79	-0.172	-0.248	-0.247	-0.06202	0.15623	0.31818
714	15	0.897	12.42	-78.00	-0.185	-0.248	-0.264	-0.06665	0.16439	0.33463
714	16	0.897	12.42	-90.00	-0.188	-0.252	-0.272	-0.06776	0.16780	0.33729
715	1	0.898	12.43	90.00	-0.185	-0.248	-0.266	-0.06692	0.16482	0.33891
715	2	0.899	12.43	78.79	-0.179	-0.241	-0.261	-0.06467	0.16124	0.33543
715	3	0.897	12.43	67.49	-0.170	-0.236	-0.250	-0.06144	0.15584	0.31973
715	4	0.896	12.42	56.29	-0.158	-0.224	-0.227	-0.05757	0.14458	0.29742
715	5	0.899	12.43	45.80	-0.153	-0.208	-0.162	-0.05538	0.13953	0.30123
715	6	0.899	12.43	33.31	-0.175	-0.172	-0.177	-0.06332	0.11912	0.32963
715	7	0.898	12.43	22.51	-0.186	-0.176	-0.179	-0.06711	0.11912	0.34218
715	8	0.897	12.43	11.00	-0.187	-0.177	-0.227	-0.06753	0.12930	0.34289
715	9	0.900	12.42	1.31	-0.172	-0.180	-0.247	-0.06253	0.13714	0.32718
715	10	0.899	12.42	12.51	-0.158	-0.190	-0.207	-0.05699	0.12898	0.30149
715	11	0.898	12.42	34.89	-0.159	-0.208	-0.194	-0.05707	0.13188	0.29605
715	12	0.899	12.41	56.29	-0.159	-0.201	-0.210	-0.05730	0.14547	0.30533
715	13	0.899	12.41	67.48	-0.170	-0.221	-0.233	-0.06134	0.14547	0.32303
715	14	0.898	12.42	78.78	-0.186	-0.239	-0.266	-0.06700	0.16195	0.33704
715	15	0.897	12.42	89.99	-0.186	-0.257	-0.274	-0.06700	0.17034	0.33594
716	1	0.897	9.31	89.99	-0.161	-0.201	-0.227	-0.05813	0.13716	0.31250
716	2	0.898	9.30	78.79	-0.149	-0.189	-0.209	-0.05394	0.12691	0.31280
716	3	0.898	9.30	67.49	-0.151	-0.198	-0.205	-0.05355	0.12606	0.30791
716	4	0.898	9.30	56.29	-0.143	-0.201	-0.191	-0.05172	0.13024	0.29793
716	5	0.898	9.30	44.99	-0.138	-0.182	-0.207	-0.04970	0.12596	0.29923
716	6	0.900	9.31	32.50	-0.146	-0.170	-0.219	-0.05266	0.12805	0.30742
716	7	0.897	9.32	11.00	-0.150	-0.160	-0.219	-0.05360	0.12151	0.30810
716	8	0.898	9.32	-1.00	-0.148	-0.141	-0.192	-0.05360	0.11065	0.30810

TABLE VA
BALANCE CAVITY AND BASE PRESSURE COEFFICIENTS, AND INCREMENTAL
AXIAL FORCE COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	M _c	α	φ	C _p c	C _p b ₁	C _p b ₂	ΔC _A c	ΔC _A b	C _A u
716	10	0.896	9.31	-11.30	-0.149	-0.139	-0.155	-0.05302	-0.09419	0.30764
716	11	0.900	9.31	-12.50	-0.136	-0.141	-0.145	-0.04980	-0.09172	0.30145
716	12	0.899	9.30	-33.80	-0.136	-0.166	-0.154	-0.04927	-0.10263	0.29605
716	13	0.899	9.30	-44.99	-0.138	-0.188	-0.152	-0.04974	-0.10854	0.29620
716	14	0.89A	9.31	-56.29	-0.140	-0.198	-0.180	-0.05209	-0.12549	0.30115
716	15	0.899	9.31	-67.79	-0.142	-0.205	-0.201	-0.05145	-0.13053	0.30503
716	17	0.900	9.31	-90.00	-0.141	-0.194	-0.199	-0.05102	-0.12557	0.30732
717	3	0.89A	6.22	-89.99	0.126	0.171	0.177	-0.04566	-0.11163	0.30161
717	4	0.899	6.22	-78.79	-0.139	-0.184	-0.175	-0.04689	-0.11517	0.30290
717	5	0.89A	6.22	-67.49	-0.131	-0.179	-0.155	-0.04677	-0.10878	0.30290
717	6	0.897	6.21	-56.29	-0.131	-0.169	-0.146	-0.04729	-0.10442	0.29801
717	7	0.89A	6.22	-44.99	-0.126	-0.154	-0.135	-0.04556	-0.09934	0.30021
717	8	0.897	6.23	-33.80	-0.131	-0.140	-0.138	-0.04738	-0.09871	0.30167
717	9	0.89A	6.23	-22.50	-0.131	-0.130	-0.136	-0.04673	-0.09660	0.30465
717	10	0.899	6.23	-11.00	-0.133	-0.132	-0.161	-0.04674	-0.09415	0.30354
717	11	0.900	6.23	-1.29	-0.125	-0.135	-0.184	-0.04875	-0.10251	0.30495
717	12	0.899	6.21	23.79	-0.125	-0.142	-0.192	-0.04650	-0.10716	0.30010
717	13	0.899	6.22	34.99	-0.134	-0.153	-0.186	-0.04851	-0.11094	0.30990
717	14	0.89A	6.20	46.29	-0.127	-0.171	-0.187	-0.04842	-0.11494	0.30170
717	15	0.897	6.20	56.79	-0.127	-0.173	-0.173	-0.04661	-0.11094	0.30837
717	16	0.89A	6.20	67.79	-0.127	-0.165	-0.158	-0.04590	-0.10394	0.30837
717	17	0.89A	6.20	78.79	-0.133	-0.168	-0.168	-0.04788	-0.11550	0.30012
717	19	0.897	6.20	89.99	-0.133	-0.168	-0.190	-0.04788	-0.11550	0.30012
71A	1	0.899	3.06	69.99	0.121	0.147	0.155	-0.04374	-0.09695	0.29651
71A	2	0.901	3.09	78.79	-0.118	-0.143	0.143	-0.04170	-0.09171	0.29651
71A	3	0.897	3.09	56.29	-0.125	-0.144	0.137	-0.04505	-0.09172	0.29456
71A	4	0.89C	3.09	44.99	-0.123	-0.145	0.140	-0.04461	-0.08941	0.29456
71A	5	0.897	3.10	33.79	-0.123	-0.137	0.139	-0.04429	-0.08708	0.29597
71A	6	0.89A	3.10	22.50	-0.122	-0.134	0.138	-0.04404	-0.08606	0.29756
71A	7	0.89A	3.10	11.00	-0.120	-0.129	0.133	-0.04376	-0.08396	0.29756
71A	8	0.89A	3.10	-1.29	-0.119	-0.120	0.131	-0.04291	-0.08331	0.29710
71A	9	0.899	3.10	-12.50	-0.125	-0.135	0.132	-0.04350	-0.08585	0.29536
71A	10	0.895	3.10	-23.79	-0.125	-0.135	0.132	-0.04350	-0.08585	0.29536
71A	11	0.899	3.10	-34.99	-0.118	-0.135	0.136	-0.04266	-0.08023	0.29421
71A	12	0.897	3.09	-46.29	-0.119	-0.145	0.136	-0.04318	-0.08253	0.29421
71A	13	0.895	3.09	-56.79	-0.119	-0.145	0.143	-0.04318	-0.08253	0.29421
71A	14	0.899	3.09	-67.49	-0.119	-0.145	0.143	-0.04318	-0.08253	0.29421
71A	15	0.897	3.09	-78.79	-0.119	-0.145	0.143	-0.04318	-0.08253	0.29421
71A	16	0.897	3.09	-89.99	-0.119	-0.145	0.143	-0.04318	-0.08253	0.29421

TABLE VA
BALANCE CAVITY AND BASE PRESSURE COEFFICIENTS, AND INCREMENTAL
AXIAL FORCE COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	M _c	α	Φ	C _{p,c}	C _{p,b1}	C _{p,b2}	ΔC _{A,c}	ΔC _{A,b}	C _{A,u}
718	0.898	3.09	-78.79	-0.117	-0.148	-0.146	-0.04238	-0.09450	0.29370
719	0.898	3.10	-89.99	-0.120	-0.149	-0.152	-0.04342	-0.09645	0.29517
719	0.897	-0.00	-90.00	-0.109	-0.128	-0.129	-0.03953	-0.08272	0.28465
719	0.89A	-0.00	-78.80	-0.112	-0.131	-0.131	-0.04042	-0.08395	0.28471
719	0.89A	-0.00	-67.50	-0.112	-0.133	-0.133	-0.04038	-0.08524	0.28528
719	0.89A	-0.00	-56.50	-0.115	-0.134	-0.135	-0.04153	-0.08646	0.28615
719	0.899	-0.00	-45.00	-0.111	-0.130	-0.129	-0.04001	-0.08322	0.28799
719	0.899	-0.00	-33.80	-0.113	-0.133	-0.132	-0.04082	-0.08518	0.28714
719	0.899	-0.00	-22.50	-0.113	-0.130	-0.132	-0.04195	-0.08531	0.28686
719	0.899	-0.00	-11.00	-0.113	-0.130	-0.132	-0.04107	-0.08417	0.28637
719	0.900	-0.00	11.29	-0.114	-0.127	-0.130	-0.04101	-0.08308	0.28679
719	0.899	-0.00	22.79	-0.113	-0.125	-0.133	-0.04074	-0.08190	0.28677
719	0.900	-0.00	34.99	-0.113	-0.122	-0.128	-0.03961	-0.08029	0.28792
719	0.901	-0.01	44.29	-0.115	-0.129	-0.134	-0.04152	-0.08441	0.28844
719	0.89A	-0.01	56.79	-0.114	-0.128	-0.133	-0.04129	-0.08382	0.28896
719	0.898	-0.01	67.49	-0.114	-0.128	-0.131	-0.04110	-0.08402	0.28858
719	0.898	-0.01	78.79	-0.114	-0.130	-0.131	-0.04110	-0.08402	0.28858
719	0.89A	-0.01	89.99	-0.116	-0.133	-0.134	-0.04204	-0.08571	0.28946
720	0.899	3.11	90.00	0.121	0.147	0.149	0.04377	0.09484	0.29352
720	0.899	3.11	78.79	0.113	0.139	0.144	0.04212	0.09096	0.29144
720	0.89A	3.10	67.50	0.120	0.136	0.140	0.04325	0.09015	0.28844
720	0.89A	3.10	56.00	0.116	0.128	0.135	0.04179	0.08573	0.28823
720	0.899	3.10	45.00	0.114	0.121	0.135	0.04115	0.08299	0.28706
720	0.897	3.09	33.50	0.117	0.120	0.136	0.04236	0.08245	0.28683
720	0.89A	3.10	22.29	0.121	0.120	0.132	0.04184	0.07893	0.28659
720	0.897	3.10	11.00	0.116	0.114	0.137	0.04194	0.08192	0.28674
720	0.897	3.10	-11.50	0.116	0.117	0.145	0.04231	0.08732	0.28587
720	0.897	3.10	-22.80	0.117	0.129	0.148	0.04198	0.08791	0.28589
720	0.897	3.10	-33.00	0.118	0.134	0.148	0.04271	0.09024	0.28912
720	0.897	3.10	-45.00	0.120	0.138	0.151	0.04280	0.09314	0.29351
720	0.899	3.11	-56.80	0.118	0.138	0.152	0.04280	0.09314	0.29351
720	0.897	3.11	-67.80	0.119	0.141	0.156	0.04280	0.09314	0.29351
720	0.897	3.11	-78.00	0.122	0.146	0.152	0.04411	0.09570	0.29353
721	0.896	3.10	-90.00	0.122	0.148	0.163	0.04400	0.09970	0.29994
721	0.897	3.10	-78.80	0.122	0.145	0.165	0.04426	0.09943	0.29905
721	0.896	3.10	-67.50	0.120	0.139	0.160	0.04344	0.09624	0.29518

TABLE VA
BALANCE CAVITY AND BASE PRESSURE COEFFICIENTS, AND INCREMENTAL
AXIAL FORCE COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	M_c	α	ϕ	C_{Pc}	C_{Pb1}	C_{Pb2}	ΔC_{Ac}	ΔC_{Ab}	C_{Au}
721	4	0.899	-3.10	-56.30	-0.115	-0.132	-0.155	0.04147	0.09212	0.29308
721	5	0.898	-3.09	-43.80	-0.117	-0.133	-0.153	0.04223	0.09177	0.29144
721	6	0.898	-3.09	-32.50	-0.113	-0.124	-0.149	0.04086	0.08558	0.28973
721	7	0.897	-3.12	-11.30	-0.113	-0.118	-0.135	0.04087	0.08134	0.28774
721	8	0.897	-3.10	11.29	-0.115	-0.119	-0.130	0.04162	0.07860	0.28867
721	9	0.898	-3.11	22.80	-0.114	-0.117	-0.133	0.04126	0.08170	0.28957
721	10	0.898	-3.09	33.80	-0.117	-0.123	-0.134	0.04124	0.08272	0.29035
721	11	0.896	-3.10	45.30	-0.117	-0.130	-0.139	0.04214	0.08648	0.29359
721	12	0.896	-3.11	56.50	-0.115	-0.136	-0.143	0.04288	0.08851	0.29556
721	13	0.898	-3.12	67.80	-0.118	-0.142	-0.147	0.04173	0.08894	0.29577
721	14	0.899	-3.12	90.00	-0.116	-0.142	-0.142	0.04184	0.09130	0.29988
722	1	0.896	-0.03	89.99	-0.114	-0.133	-0.135	0.04111	0.08587	0.29420
722	2	0.897	-0.03	78.79	-0.113	-0.130	-0.133	0.04185	0.08543	0.29458
722	3	0.896	-0.02	67.49	-0.117	-0.132	-0.137	0.04091	0.08408	0.29158
722	4	0.897	-0.02	56.29	-0.118	-0.132	-0.137	0.04277	0.08541	0.29082
722	5	0.897	-0.01	45.79	-0.116	-0.130	-0.136	0.04160	0.08546	0.29161
722	6	0.896	-0.01	32.29	-0.116	-0.131	-0.135	0.04190	0.08558	0.29198
722	7	0.897	-0.01	11.00	-0.113	-0.130	-0.132	0.04100	0.08555	0.29190
722	8	0.898	-0.01	-12.50	-0.114	-0.133	-0.136	0.04194	0.08535	0.29025
722	9	0.896	-0.01	-35.80	-0.119	-0.136	-0.136	0.04197	0.08788	0.29215
722	10	0.897	-0.02	-45.00	-0.113	-0.134	-0.135	0.04099	0.08560	0.29309
722	11	0.897	-0.02	-56.50	-0.113	-0.133	-0.138	0.04297	0.08917	0.29378
722	12	0.897	-0.02	-67.80	-0.114	-0.134	-0.132	0.04077	0.08482	0.29215
722	13	0.897	-0.02	-90.00	-0.116	-0.136	-0.132	0.04120	0.08544	0.29230
723	1	0.897	3.09	89.99	-0.118	-0.152	-0.150	0.04230	0.09647	0.30537
723	2	0.897	3.06	78.79	-0.120	-0.149	-0.146	0.04303	0.09544	0.30537
723	3	0.897	3.08	67.49	-0.119	-0.140	-0.142	0.04325	0.09847	0.30877
723	4	0.897	3.08	56.29	-0.119	-0.139	-0.132	0.04305	0.08717	0.29756
723	5	0.895	3.09	45.79	-0.123	-0.139	-0.138	0.04335	0.08736	0.29740
723	6	0.897	3.10	32.29	-0.125	-0.135	-0.138	0.04303	0.08503	0.29668
723	7	0.896	3.09	11.00	-0.124	-0.135	-0.135	0.04481	0.08661	0.30355

TABLE VA
BALANCE CAVITY AND BASE PRESSURE COEFFICIENTS, AND INCREMENTAL
AXIAL FORCE COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	M _c	α	φ	C _p c	C _p b ₁	C _p b ₂	ΔC _A c	ΔC _A b	C _A u
723	0.896	3.10	11.29	-0.123	-0.133	-0.135	0.04428	-0.08613	0.30440
723	0.897	3.09	22.49	-0.123	-0.136	-0.135	0.04445	-0.08679	0.30349
723	0.898	3.08	33.79	-0.122	-0.138	-0.136	0.04399	-0.08796	0.30546
723	0.899	3.08	44.99	-0.126	-0.146	-0.139	0.04366	-0.09139	0.30017
723	0.897	3.07	56.29	-0.123	-0.143	-0.136	0.04259	-0.08943	0.29856
723	0.897	3.07	67.49	-0.127	-0.151	-0.148	0.04463	-0.09607	0.30337
723	0.897	3.07	78.79	-0.124	-0.152	-0.166	0.04472	-0.10355	0.30400
724	0.899	6.19	89.99	-0.129	-0.165	-0.182	0.04660	-0.11919	0.30289
724	0.899	6.19	78.79	-0.130	-0.167	-0.172	0.04706	-0.11909	0.30750
724	0.898	6.19	67.49	-0.129	-0.163	-0.157	0.04654	-0.10269	0.30150
724	0.900	6.19	56.29	-0.123	-0.158	-0.160	0.04431	-0.10234	0.30453
724	0.899	6.19	44.99	-0.127	-0.169	-0.187	0.04679	-0.11014	0.30563
724	0.899	6.20	33.79	-0.127	-0.139	-0.187	0.04596	-0.10782	0.30590
724	0.901	6.21	22.49	-0.123	-0.126	-0.170	0.04459	-0.09501	0.30416
724	0.898	6.21	11.29	-0.129	-0.129	-0.158	0.04659	-0.09213	0.30953
724	0.898	6.21	-11.29	-0.128	-0.136	-0.135	0.04653	-0.08635	0.30806
724	0.899	6.20	-23.50	-0.127	-0.150	-0.135	0.04599	-0.09169	0.30850
724	0.899	6.20	-34.80	-0.129	-0.160	-0.143	0.04542	-0.09790	0.30594
724	0.898	6.20	-46.10	-0.131	-0.186	-0.158	0.04559	-0.11061	0.30334
724	0.897	6.20	-57.40	-0.128	-0.187	-0.172	0.04631	-0.11525	0.30362
724	0.899	6.20	-68.70	-0.128	-0.172	-0.178	0.04631	-0.11253	0.30332
725	0.898	9.33	-79.00	-0.143	-0.194	-0.199	0.05156	-0.12624	0.30749
725	0.899	9.33	-67.49	-0.148	-0.212	-0.205	0.05343	-0.13574	0.30829
725	0.900	9.33	-56.29	-0.140	-0.207	-0.174	0.05046	-0.12185	0.30491
725	0.898	9.33	-44.99	-0.139	-0.183	-0.143	0.05066	-0.11078	0.30207
725	0.898	9.33	-33.79	-0.139	-0.167	-0.154	0.05020	-0.10336	0.30286
725	0.897	9.33	-22.50	-0.146	-0.147	-0.147	0.05184	-0.09437	0.30425
725	0.898	9.33	-11.29	-0.148	-0.136	-0.155	0.05289	-0.09348	0.31227
725	0.899	9.33	0.00	-0.147	-0.142	-0.193	0.05353	-0.11928	0.31422
725	0.899	9.33	11.29	-0.139	-0.155	-0.220	0.05030	-0.12305	0.30549
725	0.898	9.33	22.49	-0.135	-0.166	-0.214	0.05078	-0.12554	0.30549
725	0.899	9.33	33.79	-0.135	-0.191	-0.201	0.04880	-0.12554	0.30289
725	0.901	9.33	44.99	-0.141	-0.197	-0.188	0.05090	-0.11911	0.30580
725	0.901	9.33	56.29	-0.144	-0.194	-0.177	0.05200	-0.11533	0.30580

TABLE VA
 BALANCE CAVITY AND BASE PRESSURE COEFFICIENTS, AND INCREMENTAL
 AXIAL FORCE COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	M _c	α	φ	C _p c	C _p b ₁	C _p b ₂	ΔC _A c	ΔC _A b	C _A u
725	0.897	9.31	78.79	-0.153	-0.196	-0.201	-0.05525	-0.12740	0.30002
725	0.89A	9.31	89.99	-0.149	-0.192	-0.217	-0.05389	-0.13104	0.30617
726	0.899	12.43	89.99	-0.185	-0.257	-0.274	-0.06683	-0.17046	0.33988
726	0.901	12.43	78.48	-0.181	-0.234	-0.261	-0.06520	-0.15855	0.33804
726	0.901	12.43	67.48	-0.165	-0.216	-0.227	-0.05975	-0.14209	0.32861
726	0.900	12.43	56.29	-0.152	-0.196	-0.205	-0.05507	-0.12876	0.30239
726	0.89A	12.44	44.80	-0.161	-0.214	-0.195	-0.05810	-0.15116	0.30867
726	0.900	12.44	33.80	-0.153	-0.183	-0.201	-0.05539	-0.12327	0.30416
726	0.89A	12.44	22.51	-0.173	-0.183	-0.250	-0.06261	-0.15882	0.33119
726	0.89A	12.44	11.00	-0.186	-0.179	-0.269	-0.06798	-0.14375	0.34479
726	0.89A	12.45	11.00	-0.188	-0.193	-0.228	-0.06718	-0.13088	0.34542
726	0.89A	12.45	-11.51	-0.186	-0.173	-0.177	-0.06231	-0.11227	0.33524
726	0.89A	12.45	-33.80	-0.156	-0.150	-0.166	-0.05617	-0.10149	0.30504
726	0.897	12.45	-44.29	-0.156	-0.169	-0.159	-0.05604	-0.11355	0.30924
726	0.897	12.46	-56.29	-0.167	-0.232	-0.212	-0.06502	-0.15402	0.30180
726	0.899	12.46	-78.79	-0.181	-0.240	-0.266	-0.06551	-0.16221	0.32080
726	0.900	12.46	-90.00	-0.185	-0.249	-0.261	-0.06685	-0.16358	0.33434
727	0.89A	12.45	90.01	-0.184	-0.245	-0.256	-0.06700	-0.16068	0.34984
727	0.897	12.45	-78.49	-0.172	-0.238	-0.268	-0.06652	-0.16233	0.34519
727	0.896	12.45	-56.29	-0.159	-0.235	-0.252	-0.06202	-0.15393	0.32959
727	0.896	12.44	-44.80	-0.157	-0.225	-0.224	-0.05738	-0.14397	0.31362
727	0.897	12.44	-33.80	-0.161	-0.155	-0.165	-0.05632	-0.10606	0.31362
727	0.897	12.44	-22.51	-0.179	-0.177	-0.183	-0.05817	-0.11566	0.34124
727	0.89A	12.44	11.00	-0.182	-0.189	-0.177	-0.06465	-0.11746	0.35732
727	0.89A	12.44	11.00	-0.187	-0.178	-0.225	-0.06660	-0.12869	0.35732
727	0.897	12.42	33.80	-0.176	-0.185	-0.269	-0.06736	-0.14345	0.35338
727	0.899	12.42	44.29	-0.156	-0.187	-0.254	-0.06355	-0.14075	0.35134
727	0.897	12.42	56.29	-0.159	-0.201	-0.191	-0.05631	-0.12706	0.31066
727	0.898	12.41	67.48	-0.159	-0.201	-0.210	-0.05727	-0.13206	0.31066
727	0.897	12.42	78.78	-0.172	-0.223	-0.237	-0.06210	-0.14745	0.33474
727	0.896	12.42	89.99	-0.189	-0.236	-0.267	-0.06643	-0.16126	0.34714
72A	0.89A	9.30	89.99	-0.152	-0.192	-0.219	-0.05472	-0.13177	0.32513
72A	0.89A	9.30	78.79	-0.150	-0.189	-0.208	-0.05422	-0.12743	0.32143
72A	0.897	9.30	67.49	-0.142	-0.193	-0.177	-0.05139	-0.11806	0.31301

TABLE VA
 BALANCE CAVITY AND BASE PRESSURE COEFFICIENTS, AND INCREMENTAL
 AXIAL FORCE COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	M _c	α	φ	C _p c	C _p b ₁	C _p b ₂	ΔC _A c	ΔC _A b	C _A u
728	4	0.897	9.30	56.29	-0.138	-0.197	-0.188	0.05000	-0.12348	0.31160
72A	5	0.898	9.30	44.79	-0.136	-0.192	-0.202	0.04925	-0.12636	0.30912
72A	6	0.897	9.30	33.79	-0.137	-0.182	-0.212	0.04938	-0.12612	0.30976
72A	7	0.897	9.30	22.50	-0.142	-0.165	-0.225	0.05137	-0.12519	0.31934
72A	8	0.895	9.31	11.30	-0.148	-0.158	-0.218	0.05350	-0.12072	0.31935
72A	9	0.896	9.32	-0.00	-0.148	-0.143	-0.196	0.05358	-0.10890	0.32243
72A	10	0.897	9.32	-11.30	-0.145	-0.137	-0.155	0.05243	-0.09368	0.31145
72A	11	0.896	9.31	-23.79	-0.141	-0.146	-0.150	0.05076	-0.09515	0.31155
72A	12	0.897	9.31	-33.79	-0.135	-0.166	-0.151	0.04885	-0.10172	0.31013
72A	13	0.896	9.31	-44.29	-0.140	-0.169	-0.146	0.05065	-0.10774	0.31033
72A	14	0.89A	9.32	-56.29	-0.140	-0.198	-0.152	0.05043	-0.11229	0.31039
72A	15	0.897	9.32	-67.79	-0.142	-0.208	-0.174	0.05112	-0.12251	0.31154
72A	16	0.898	9.32	-78.79	-0.144	-0.208	-0.202	0.05191	-0.13154	0.31540
72A	17	0.897	9.32	-89.99	-0.150	-0.206	-0.206	0.05419	-0.13230	0.32124
729	1	0.897	6.21	89.99	-0.126	-0.176	-0.181	0.04537	-0.11488	0.31002
729	2	0.89A	6.21	78.79	-0.126	-0.187	-0.168	0.04556	-0.11398	0.31021
729	3	0.896	6.21	-67.79	-0.130	-0.188	-0.156	0.04715	-0.11055	0.31756
729	4	0.900	6.21	-56.29	-0.121	-0.174	-0.139	0.04391	-0.10093	0.31055
729	5	0.897	6.21	-43.79	-0.128	-0.172	-0.143	0.04638	-0.10093	0.31033
729	6	0.898	6.22	-32.79	-0.130	-0.156	-0.138	0.04684	-0.09941	0.31505
729	7	0.897	6.22	-22.30	-0.129	-0.141	-0.140	0.04679	-0.08731	0.31509
729	8	0.898	6.22	-11.00	-0.126	-0.132	-0.140	0.04547	-0.08088	0.31700
729	9	0.89A	6.21	11.29	-0.128	-0.132	-0.177	0.04611	-0.09921	0.31610
729	10	0.89A	6.21	23.79	-0.125	-0.136	-0.184	0.04522	-0.10924	0.31293
729	11	0.897	6.20	35.79	-0.129	-0.152	-0.189	0.04674	-0.10942	0.31293
729	12	0.896	6.20	44.29	-0.128	-0.167	-0.183	0.04619	-0.11026	0.31100
729	13	0.897	6.20	56.29	-0.128	-0.164	-0.169	0.04681	-0.10768	0.31087
729	14	0.897	6.21	67.79	-0.128	-0.164	-0.157	0.04695	-0.10377	0.31025
729	15	0.89A	6.20	78.79	-0.130	-0.168	-0.177	0.04681	-0.11055	0.31253
729	16	0.89A	6.20	89.99	-0.126	-0.164	-0.180	0.04567	-0.11055	0.31081
729	17	0.898	6.21	89.99	-0.126	-0.164	-0.180	0.04567	-0.11055	0.31081
730	1	0.897	3.08	89.99	-0.125	-0.155	-0.171	0.04521	-0.10459	0.31057
730	2	0.898	3.07	78.79	-0.120	-0.149	-0.162	0.04338	-0.10991	0.31200
730	3	0.898	3.07	67.49	-0.121	-0.151	-0.156	0.04365	-0.09867	0.31300
730	4	0.900	3.08	56.29	-0.123	-0.150	-0.140	0.04228	-0.09594	0.31247
730	5	0.900	3.08	44.29	-0.120	-0.145	-0.138	0.04351	-0.08999	0.31074
730	6	0.89A	3.09	33.79	-0.127	-0.145	-0.143	0.04576	-0.09239	0.30951
730	7	0.89A	3.10	22.49	-0.127	-0.141	-0.141	0.04581	-0.09051	0.31111
730	8	0.89A	3.10	11.00	-0.125	-0.137	-0.136	0.04505	-0.08756	0.31110
730	9	0.89A	3.11	-0.00	-0.126	-0.137	-0.137	0.04537	-0.08812	0.31110

TABLE VA
BALANCE CAVITY AND BASE PRESSURE COEFFICIENTS, AND INCREMENTAL
AXIAL FORCE COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	M _c	α	φ	C _{p,c}	C _{p,b1}	C _{p,b2}	ΔC _{A,c}	ΔC _{A,b}	C _{A,u}
730	10	0.897	3.10	-11.30	-0.125	-0.136	-0.135	-0.04512	-0.08731	0.30952
730	11	0.899	3.10	-22.50	-0.122	-0.134	-0.130	-0.04399	-0.08459	0.30716
730	12	0.898	3.10	-33.79	-0.119	-0.136	-0.130	-0.04310	-0.08529	0.30396
730	13	0.898	3.10	-44.99	-0.117	-0.139	-0.131	-0.04228	-0.08663	0.30577
730	14	0.897	3.11	-56.29	-0.122	-0.148	-0.140	-0.04223	-0.09261	0.30517
730	15	0.89A	3.09	-67.49	-0.120	-0.148	-0.148	-0.04347	-0.09636	0.30591
730	16	0.89A	3.10	-78.79	-0.120	-0.153	-0.152	-0.04334	-0.09793	0.30733
730	17	0.900	3.10	-89.99	-0.120	-0.153	-0.152	-0.04334	-0.09793	0.30733
731	1	0.89A	0.00	90.00	-0.117	0.138	-0.136	-0.04232	-0.08791	0.29916
731	2	0.900	0.00	-78.80	-0.114	0.134	-0.134	-0.04117	-0.08618	0.30127
731	3	0.900	0.00	-67.50	-0.114	0.134	-0.135	-0.04136	-0.08669	0.30126
731	4	0.898	0.00	-56.30	-0.118	0.139	-0.141	-0.04280	-0.09038	0.30356
731	5	0.899	0.00	-45.00	-0.119	0.136	-0.139	-0.04290	-0.08928	0.30386
731	6	0.900	0.01	-33.80	-0.117	0.137	-0.138	-0.04225	-0.08843	0.30296
731	7	0.900	0.01	-22.50	-0.118	0.130	-0.135	-0.04253	-0.08836	0.30052
731	8	0.901	0.02	-11.00	-0.112	0.129	-0.135	-0.04370	-0.08454	0.29944
731	9	0.899	0.02	0.00	-0.113	0.128	-0.135	-0.04331	-0.08465	0.29989
731	10	0.897	0.01	12.49	-0.120	0.135	-0.144	-0.04375	-0.08940	0.29909
731	11	0.89A	0.01	23.79	-0.120	0.134	-0.147	-0.04351	-0.08834	0.30042
731	12	0.900	0.00	34.99	-0.116	0.132	-0.141	-0.04298	-0.08617	0.30216
731	13	0.898	0.00	46.29	-0.117	0.134	-0.138	-0.04271	-0.08723	0.30139
731	14	0.898	0.00	57.49	-0.117	0.136	-0.140	-0.04271	-0.08723	0.30139
731	15	0.899	0.00	67.79	-0.117	0.136	-0.140	-0.04271	-0.08723	0.30139
731	16	0.899	0.00	78.99	-0.117	0.136	-0.140	-0.04271	-0.08723	0.30139
731	17	0.899	0.00	89.99	-0.117	0.136	-0.140	-0.04271	-0.08723	0.30139
732	1	0.899	3.11	90.00	-0.118	0.149	-0.148	-0.04258	-0.09517	0.30405
732	2	0.89A	3.11	78.80	-0.117	0.143	-0.147	-0.04241	-0.09341	0.30973
732	3	0.900	3.10	67.50	-0.115	0.139	-0.147	-0.04268	-0.09159	0.30890
732	4	0.899	3.10	56.30	-0.115	0.132	-0.141	-0.04261	-0.08703	0.29659
732	5	0.899	3.09	45.00	-0.116	0.125	-0.138	-0.04220	-0.08470	0.29459
732	6	0.900	3.08	33.80	-0.117	0.118	-0.135	-0.04231	-0.08217	0.29321
732	7	0.900	3.08	22.50	-0.119	0.118	-0.137	-0.04289	-0.08217	0.29321
732	8	0.900	3.08	11.00	-0.115	0.122	-0.136	-0.04170	-0.08182	0.29330
732	9	0.898	3.07	0.00	-0.113	0.128	-0.151	-0.04194	-0.08962	0.29341
732	10	0.898	3.08	-11.30	-0.116	0.131	-0.158	-0.04235	-0.09296	0.29674
732	11	0.899	3.08	-22.50	-0.117	0.138	-0.160	-0.04355	-0.09441	0.29746
732	12	0.899	3.08	-33.80	-0.120	0.141	-0.162	-0.04344	-0.09637	0.30269
732	13	0.89A	3.09	-45.00	-0.120	0.141	-0.162	-0.04344	-0.09637	0.30269
732	14	0.897	3.09	-56.30	-0.120	0.141	-0.162	-0.04344	-0.09637	0.30269
732	15	0.897	3.09	-67.50	-0.120	0.141	-0.162	-0.04344	-0.09637	0.30269
732	16	0.899	3.09	-78.80	-0.120	0.141	-0.162	-0.04344	-0.09637	0.30269
732	17	0.899	3.09	-89.99	-0.120	0.141	-0.162	-0.04344	-0.09637	0.30269

TABLE VA
BALANCE CAVITY AND BASE PRESSURE COEFFICIENTS, AND INCREMENTAL AXIAL FORCE COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	M_c	α	ϕ	C_{p_c}	$C_{p_{b1}}$	$C_{p_{b2}}$	ΔC_{A_c}	ΔC_{A_b}	C_{A_u}
732	17	0.898	-3.09	-90.00	-0.123	-0.147	-0.165	-0.044	-0.100	0.300
733	1	0.899	-3.09	90.00	-0.120	-0.145	-0.161	-0.043	-0.098	0.320
733	2	0.899	-3.08	78.80	-0.118	-0.140	-0.160	-0.042	-0.097	0.316
733	3	0.899	-3.06	-67.30	-0.119	-0.137	-0.159	-0.042	-0.095	0.307
733	4	0.900	-3.06	56.00	-0.118	-0.134	-0.159	-0.042	-0.094	0.306
733	5	0.900	-3.05	-45.00	-0.112	-0.128	-0.156	-0.041	-0.090	0.307
733	6	0.898	-3.05	33.80	-0.114	-0.130	-0.156	-0.041	-0.091	0.304
733	7	0.899	-3.04	-22.30	-0.114	-0.127	-0.149	-0.041	-0.088	0.305
733	8	0.898	-3.04	11.00	-0.114	-0.121	-0.142	-0.041	-0.086	0.305
733	9	0.898	-3.04	-10.29	-0.115	-0.120	-0.139	-0.041	-0.083	0.306
733	10	0.898	-3.05	12.29	-0.115	-0.118	-0.137	-0.041	-0.081	0.306
733	11	0.899	-3.05	33.79	-0.116	-0.115	-0.139	-0.041	-0.084	0.307
733	12	0.898	-3.06	45.00	-0.117	-0.130	-0.143	-0.041	-0.087	0.309
733	13	0.900	-3.07	56.30	-0.116	-0.133	-0.144	-0.041	-0.090	0.312
733	14	0.898	-3.08	67.80	-0.116	-0.139	-0.144	-0.041	-0.090	0.312
733	15	0.896	-3.09	78.80	-0.120	-0.150	-0.150	-0.041	-0.096	0.315
733	16	0.898	-3.10	90.00	-0.116	-0.148	-0.147	-0.041	-0.094	0.317
734	1	0.899	0.01	89.99	-0.116	-0.136	-0.148	-0.041	-0.091	0.315
734	2	0.897	0.02	78.79	-0.119	-0.138	-0.151	-0.041	-0.092	0.314
734	3	0.898	0.03	67.49	-0.120	-0.137	-0.151	-0.041	-0.092	0.314
734	4	0.898	0.03	56.29	-0.122	-0.136	-0.153	-0.041	-0.092	0.314
734	5	0.896	0.04	44.99	-0.116	-0.133	-0.153	-0.041	-0.092	0.312
734	6	0.897	0.05	33.49	-0.117	-0.133	-0.150	-0.041	-0.090	0.313
734	7	0.898	0.06	22.29	-0.114	-0.131	-0.146	-0.041	-0.088	0.313
734	8	0.897	0.06	11.00	-0.115	-0.131	-0.146	-0.041	-0.088	0.313
734	9	0.898	0.06	0.00	-0.120	-0.135	-0.150	-0.041	-0.091	0.312
734	10	0.898	0.06	-11.50	-0.117	-0.138	-0.147	-0.041	-0.091	0.314
734	11	0.896	0.04	23.80	-0.118	-0.140	-0.149	-0.041	-0.092	0.315
734	12	0.897	0.05	35.00	-0.121	-0.142	-0.152	-0.041	-0.092	0.315
734	13	0.897	0.05	46.30	-0.120	-0.139	-0.149	-0.041	-0.092	0.317
734	14	0.898	0.03	57.50	-0.124	-0.141	-0.152	-0.041	-0.094	0.315
734	15	0.897	0.03	67.80	-0.121	-0.140	-0.148	-0.041	-0.092	0.315
734	16	0.896	0.03	78.80	-0.121	-0.140	-0.148	-0.041	-0.092	0.315
734	17	0.896	0.03	90.00	-0.121	-0.140	-0.148	-0.041	-0.092	0.315
735	1	0.896	3.16	89.99	-0.121	-0.157	-0.154	-0.041	-0.099	0.320
735	2	0.897	3.16	78.79	-0.119	-0.154	-0.155	-0.041	-0.095	0.319
735	3	0.898	3.16	67.49	-0.121	-0.150	-0.153	-0.041	-0.092	0.317
735	4	0.896	3.17	56.29	-0.122	-0.147	-0.155	-0.041	-0.090	0.315

TABLE VA
BALANCE CAVITY AND BASE PRESSURE COEFFICIENTS, AND INCREMENTAL AXIAL FORCE COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	M _c	α	φ	C _p c	C _p b ₁	C _p b ₂	ΔC _A c	ΔC _A b	C _A u
735	5	0.895	3.17	44.99	-0.123	-0.146	-0.136	0.0460	0.09032	0.31535
735	6	0.896	3.17	33.79	-0.116	-0.134	-0.127	0.04190	0.08370	0.31612
735	7	0.897	3.17	22.50	-0.118	-0.133	-0.129	0.04268	0.08419	0.31715
735	8	0.897	3.17	11.30	-0.123	-0.134	-0.133	0.04329	0.08582	0.32075
735	9	0.897	3.17	0.00	-0.122	-0.132	-0.133	0.04362	0.08496	0.32062
735	10	0.897	3.16	11.29	-0.127	-0.141	-0.137	0.04419	0.08775	0.31735
735	11	0.896	3.16	22.49	-0.121	-0.139	-0.136	0.04376	0.08820	0.31735
735	12	0.898	3.15	33.79	-0.121	-0.147	-0.133	0.04384	0.08983	0.32240
735	13	0.898	3.15	44.29	-0.120	-0.147	-0.144	0.04347	0.09321	0.32405
735	14	0.898	3.14	56.29	-0.121	-0.152	-0.163	0.04385	0.10118	0.32506
735	15	0.898	3.14	67.49	-0.121	-0.153	-0.167	0.04522	0.10268	0.32531
735	16	0.895	3.14	78.79	-0.125	-0.145	-0.160	0.04297	0.10979	0.32540
735	17	0.897	3.14	89.99	-0.119	-0.145	-0.160	0.04297	0.10979	0.32540
736	1	0.897	6.26	89.99	-0.129	-0.165	-0.182	0.04657	0.1146	0.31740
736	2	0.896	6.26	78.79	-0.131	-0.168	-0.175	0.04736	0.11019	0.32239
736	3	0.896	6.26	67.49	-0.128	-0.166	-0.158	0.04620	0.10379	0.32154
736	4	0.899	6.26	56.29	-0.128	-0.156	-0.164	0.04601	0.10356	0.32196
736	5	0.896	6.26	44.29	-0.129	-0.156	-0.179	0.04618	0.10765	0.32123
736	6	0.900	6.27	33.79	-0.127	-0.137	-0.180	0.04663	0.10768	0.32262
736	7	0.897	6.27	22.49	-0.128	-0.131	-0.171	0.04618	0.10178	0.32262
736	8	0.896	6.27	11.00	-0.130	-0.126	-0.155	0.04706	0.09023	0.32262
736	9	0.897	6.27	0.00	-0.131	-0.138	-0.138	0.04737	0.08718	0.32262
736	10	0.897	6.27	11.50	-0.129	-0.138	-0.134	0.04657	0.08718	0.32262
736	11	0.897	6.27	22.79	-0.129	-0.155	-0.138	0.04657	0.09415	0.32503
736	12	0.897	6.27	33.99	-0.130	-0.174	-0.134	0.04704	0.10179	0.32503
736	13	0.897	6.27	44.29	-0.127	-0.177	-0.143	0.04598	0.10202	0.32503
736	14	0.898	6.27	56.29	-0.129	-0.183	-0.148	0.04598	0.10202	0.32503
736	15	0.897	6.27	67.49	-0.127	-0.183	-0.148	0.04597	0.10278	0.32503
736	16	0.897	6.27	78.79	-0.127	-0.183	-0.163	0.04597	0.11278	0.32503
736	17	0.897	6.27	89.99	-0.131	-0.183	-0.179	0.04744	0.11604	0.32503
737	1	0.896	9.41	90.00	-0.151	-0.206	-0.204	0.05458	0.13154	0.33019
737	2	0.897	9.42	78.79	-0.145	-0.212	-0.198	0.05331	0.13520	0.33019
737	3	0.898	9.41	67.48	-0.138	-0.212	-0.166	0.05233	0.12109	0.33201
737	4	0.899	9.41	56.29	-0.137	-0.195	-0.149	0.04990	0.10360	0.33201
737	5	0.897	9.41	44.29	-0.137	-0.182	-0.138	0.04953	0.10360	0.33201
737	6	0.898	9.41	33.79	-0.137	-0.162	-0.149	0.04957	0.10000	0.33201
737	7	0.898	9.41	22.50	-0.137	-0.141	-0.144	0.04961	0.09152	0.33201
737	8	0.898	9.41	11.30	-0.145	-0.134	-0.157	0.05189	0.09356	0.33201
737	9	0.898	9.41	0.00	-0.145	-0.141	-0.193	0.05231	0.10711	0.33201
737	10	0.898	9.41	11.30	-0.146	-0.154	-0.218	0.05275	0.11936	0.33201

TABLE VA
BALANCE CAVITY AND BASE PRESSURE COEFFICIENTS, AND INCREMENTAL AXIAL FORCE COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	M _c	α	φ	C _p c	C _p b ₁	C _p b ₂	ΔC _A c	ΔC _A b	C _A u
737	11	0.897	9.40	22.50	-0.141	-0.165	-0.222	-0.05078	-0.12434	0.31803
737	12	0.89A	9.39	33.79	-0.139	-0.162	-0.214	-0.05006	-0.12686	0.31658
737	13	0.89A	9.39	44.29	-0.142	-0.199	-0.207	-0.05107	-0.12930	0.31466
737	14	0.89A	9.39	56.48	-0.140	-0.190	-0.191	-0.05066	-0.12716	0.31644
737	15	0.897	9.39	78.79	-0.147	-0.192	-0.195	-0.05334	-0.12403	0.31241
737	17	0.897	9.39	89.99	-0.148	-0.191	-0.217	-0.05334	-0.13060	0.32095
73A	1	0.897	12.50	89.99	-0.185	-0.256	-0.273	-0.06681	-0.16960	0.35536
73A	2	0.897	12.50	78.78	-0.173	-0.240	-0.271	-0.06733	-0.16371	0.35455
73A	3	0.897	12.49	67.48	-0.158	-0.225	-0.238	-0.06208	-0.14841	0.35816
73A	4	0.899	12.49	56.29	-0.154	-0.207	-0.209	-0.05708	-0.13137	0.31767
73A	5	0.899	12.50	44.29	-0.161	-0.193	-0.189	-0.05548	-0.12980	0.31698
73A	6	0.899	12.51	33.79	-0.174	-0.184	-0.212	-0.05811	-0.14328	0.34478
73B	7	0.899	12.51	22.51	-0.184	-0.178	-0.256	-0.06649	-0.14328	0.36025
73A	8	0.897	12.52	11.00	-0.191	-0.183	-0.252	-0.06899	-0.13307	0.36211
73A	9	0.897	12.51	0.30	-0.187	-0.197	-0.182	-0.06764	-0.12171	0.36194
73A	10	0.898	12.51	-12.51	-0.176	-0.180	-0.184	-0.06367	-0.11607	0.32456
73A	11	0.901	12.52	-23.79	-0.153	-0.146	-0.169	-0.05509	-0.10720	0.31911
73A	12	0.897	12.52	-34.99	-0.158	-0.192	-0.173	-0.05714	-0.11720	0.31250
73A	13	0.898	12.52	-46.28	-0.155	-0.214	-0.204	-0.05583	-0.13415	0.34357
73A	14	0.901	12.53	-57.48	-0.166	-0.239	-0.243	-0.05984	-0.15155	0.35421
73A	15	0.898	12.54	-78.68	-0.162	-0.243	-0.258	-0.06505	-0.15398	0.35421
73A	16	0.898	12.53	-90.01	-0.162	-0.243	-0.251	-0.06580	-0.15829	0.35601
739	3	0.799	3.12	0.00	-0.114	-0.114	-0.127	-0.04124	-0.07726	0.29371
739	4	0.800	-1.06	0.00	-0.107	-0.116	-0.129	-0.03860	-0.07878	0.28718
739	5	0.800	0.03	0.00	-0.105	-0.113	-0.126	-0.03804	-0.08013	0.28407
739	6	0.800	0.48	0.00	-0.106	-0.126	-0.123	-0.03834	-0.08027	0.28407
739	7	0.799	0.97	0.00	-0.106	-0.127	-0.121	-0.03840	-0.08066	0.28628
739	8	0.798	3.10	0.00	-0.108	-0.124	-0.126	-0.03906	-0.08034	0.28467
739	9	0.798	6.20	0.00	-0.119	-0.138	-0.157	-0.04315	-0.09034	0.28687
739	10	0.797	9.30	0.00	-0.143	-0.138	-0.181	-0.05155	-0.10249	0.30052
739	11	0.797	12.51	0.00	-0.174	-0.175	-0.214	-0.06281	-0.12480	0.31752
739	12	0.800	-0.07	0.00	-0.104	-0.122	-0.127	-0.03759	-0.08013	0.28172
740	1	0.79A	3.13	0.00	-0.109	-0.111	-0.123	-0.03925	-0.07519	0.28552
740	2	0.79A	-1.06	0.00	-0.105	-0.119	-0.127	-0.03811	-0.07898	0.28282
740	3	0.799	0.02	0.00	-0.105	-0.121	-0.124	-0.03842	-0.07872	0.28151
740	4	0.797	0.49	0.00	-0.106	-0.126	-0.123	-0.03848	-0.08026	0.28198
740	5	0.797	0.98	0.00	-0.108	-0.128	-0.124	-0.03897	-0.08115	0.28455

TABLE VA
 BALANCE CAVITY AND BASE PRESSURE COEFFICIENTS, AND INCREMENTAL
 AXIAL FORCE COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	M_c	α	Φ	C_{p_c}	$C_{p_{b1}}$	$C_{p_{b2}}$	ΔC_{A_c}	ΔC_{A_b}	C_{A_u}
740	6	0.799	3.09	-0.00	-0.105	-0.119	-0.125	-0.03791	-0.07046	0.20507
740	7	0.79A	6.22	-0.00	-0.115	-0.120	-0.153	-0.04168	-0.08772	0.20059
740	8	0.79A	9.32	-0.00	-0.143	-0.140	-0.104	-0.05174	-0.10403	0.30421
740	9	0.797	12.50	-0.00	-0.175	-0.176	-0.212	-0.06317	-0.12436	0.32167
740	10	0.797	-0.04	-0.00	-0.108	-0.125	-0.126	-0.03911	-0.08078	0.20164
741	1	0.79A	-3.10	-0.00	-0.107	-0.112	-0.122	-0.03862	-0.07534	0.20319
741	2	0.797	-1.01	-0.00	-0.107	-0.119	-0.129	-0.03867	-0.07981	0.20226
741	3	0.797	-0.01	-0.00	-0.106	-0.124	-0.125	-0.03847	-0.07977	0.20004
741	4	0.797	0.49	-0.00	-0.109	-0.129	-0.125	-0.03924	-0.08147	0.20179
741	5	0.79A	1.01	-0.00	-0.103	-0.124	-0.119	-0.03735	-0.07008	0.20353
741	6	0.797	3.10	-0.00	-0.110	-0.124	-0.127	-0.03975	-0.08089	0.20609
741	7	0.799	6.25	-0.00	-0.114	-0.116	-0.148	-0.04121	-0.08469	0.20907
741	8	0.79A	9.35	-0.00	-0.142	-0.139	-0.103	-0.05140	-0.10313	0.30577
741	9	0.797	12.49	-0.00	-0.171	-0.171	-0.214	-0.06176	-0.12323	0.32237
741	10	0.797	-0.02	-0.00	-0.103	-0.120	-0.123	-0.03732	-0.07810	0.27985
742	1	0.798	-3.10	-0.00	-0.108	-0.110	-0.125	-0.03915	-0.07564	0.20007
742	2	0.798	-1.08	-0.00	-0.102	-0.114	-0.125	-0.03699	-0.07673	0.20194
742	3	0.79C	-0.03	-0.00	-0.108	-0.125	-0.128	-0.03889	-0.08149	0.20128
742	4	0.796	0.51	-0.00	-0.110	-0.130	-0.126	-0.03970	-0.08209	0.20187
742	5	0.798	1.00	-0.00	-0.104	-0.123	-0.120	-0.03768	-0.07817	0.20342
742	6	0.797	3.15	-0.00	-0.111	-0.124	-0.130	-0.04012	-0.08179	0.20048
742	7	0.79A	6.23	-0.00	-0.116	-0.119	-0.152	-0.04198	-0.08718	0.29402
742	8	0.797	9.33	-0.00	-0.145	-0.138	-0.183	-0.05230	-0.10308	0.30790
742	9	0.797	12.50	-0.00	-0.174	-0.173	-0.217	-0.06282	-0.12493	0.32648
742	10	0.800	-0.01	-0.00	-0.101	-0.120	-0.119	-0.03649	-0.07673	0.27922
743	1	0.799	-3.08	-0.00	-0.104	-0.107	-0.125	-0.03776	-0.07470	0.27050
743	2	0.799	-1.03	-0.00	-0.104	-0.118	-0.128	-0.03744	-0.07899	0.20231
743	3	0.79A	-0.02	-0.00	-0.106	-0.127	-0.123	-0.03826	-0.08037	0.20198
743	4	0.799	0.50	-0.00	-0.107	-0.125	-0.120	-0.03857	-0.07960	0.20303
743	5	0.799	1.04	-0.00	-0.107	-0.126	-0.122	-0.03856	-0.07975	0.20655
743	6	0.799	3.14	-0.00	-0.111	-0.120	-0.129	-0.04002	-0.08012	0.29587
743	7	0.799	6.24	-0.00	-0.121	-0.120	-0.153	-0.04378	-0.08792	0.30054
743	8	0.799	9.35	-0.00	-0.147	-0.141	-0.187	-0.05318	-0.10547	0.31328
743	9	0.783	12.50	-0.00	-0.169	-0.167	-0.209	-0.06100	-0.12063	0.32988
743	10	0.799	-0.01	-0.00	-0.106	-0.128	-0.124	-0.0383A	-0.08112	0.20117
744	1	0.800	-3.05	-0.00	-0.103	-0.109	-0.123	-0.03726	-0.07450	0.27995
744	2	0.799	-1.01	-0.00	-0.106	-0.127	-0.133	-0.03849	-0.08348	0.20796
744	3	0.799	-0.00	-0.00	-0.103	-0.127	-0.124	-0.03735	-0.08088	0.20992

TABLE VA
BALANCE CAVITY AND BASE PRESSURE COEFFICIENTS, AND INCREMENTAL
AXIAL FORCE COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	M _c	α	Φ	C _p c	C _p b _l	C _p b ₂	ΔC _A c	ΔC _A b	C _A u
744	0.799	0.50	-0.00	-0.108	-0.133	-0.125	-0.03922	-0.08293	0.29250
744	0.800	1.05	-0.00	-0.110	-0.132	-0.125	-0.03973	-0.08254	0.29435
744	0.799	3.13	-0.00	-0.116	-0.125	-0.133	-0.04182	-0.08274	0.30594
744	0.799	6.26	-0.00	-0.123	-0.122	-0.154	-0.04453	-0.08881	0.31062
744	0.799	9.34	-0.00	-0.142	-0.135	-0.182	-0.05120	-0.10191	0.32219
744	0.800	12.53	-0.00	-0.167	-0.165	-0.206	-0.06036	-0.11931	0.33689
744	0.800	0.01	-0.00	-0.107	-0.131	-0.126	-0.03852	-0.08254	0.28896
745	0.801	-3.04	-0.00	-0.104	-0.110	-0.128	-0.03744	-0.07640	0.28666
745	0.800	-1.01	-0.00	-0.105	-0.129	-0.135	-0.03789	-0.08483	0.30019
745	0.800	0.02	-0.00	-0.107	-0.140	-0.134	-0.03868	-0.08773	0.30261
745	0.800	0.58	-0.00	-0.106	-0.137	-0.126	-0.03851	-0.08443	0.30497
745	0.800	1.08	-0.00	-0.109	-0.135	-0.124	-0.03928	-0.08319	0.30447
745	0.800	3.17	-0.00	-0.113	-0.125	-0.135	-0.04088	-0.08349	0.31717
745	0.800	6.26	-0.00	-0.121	-0.120	-0.152	-0.04391	-0.08724	0.32963
745	0.801	9.36	-0.00	-0.140	-0.134	-0.180	-0.05071	-0.10108	0.34331
745	0.800	12.54	-0.00	-0.167	-0.164	-0.211	-0.06040	-0.12018	0.35067
745	0.800	0.03	-0.00	-0.107	-0.138	-0.133	-0.03866	-0.08724	0.30067
746	0.801	-3.01	-0.00	-0.105	-0.116	-0.134	-0.03792	-0.08013	0.29805
746	0.800	-0.99	-0.00	-0.102	-0.132	-0.140	-0.03690	-0.08724	0.31204
746	0.798	0.02	-0.00	-0.107	-0.142	-0.137	-0.03868	-0.08978	0.31552
746	0.799	0.53	-0.00	-0.104	-0.137	-0.128	-0.03767	-0.08521	0.31822
746	0.798	1.11	-0.00	-0.107	-0.137	-0.124	-0.03853	-0.08400	0.31907
746	0.800	3.18	-0.00	-0.117	-0.132	-0.147	-0.04226	-0.08754	0.33035
746	0.800	6.26	-0.00	-0.116	-0.135	-0.147	-0.04192	-0.08431	0.33593
746	0.799	9.33	-0.00	-0.140	-0.137	-0.181	-0.05054	-0.10193	0.33593
746	0.800	12.54	-0.00	-0.166	-0.164	-0.210	-0.06000	-0.12012	0.35003
746	0.800	0.07	-0.00	-0.104	-0.140	-0.135	-0.03760	-0.08833	0.31477
747	0.799	-2.98	-0.00	-0.107	-0.118	-0.138	-0.03883	-0.08215	0.31090
747	0.800	-0.97	-0.00	-0.102	-0.132	-0.142	-0.03707	-0.08860	0.32630
747	0.799	0.04	-0.00	-0.108	-0.144	-0.142	-0.03891	-0.09157	0.33090
747	0.799	0.56	-0.00	-0.105	-0.144	-0.135	-0.03803	-0.08941	0.33273
747	0.801	1.09	-0.00	-0.104	-0.136	-0.125	-0.03779	-0.08362	0.34047
747	0.799	3.18	-0.00	-0.118	-0.135	-0.148	-0.04258	-0.09002	0.35975
747	0.800	6.27	-0.00	-0.118	-0.117	-0.178	-0.04988	-0.09988	0.35975
747	0.799	9.37	-0.00	-0.138	-0.133	-0.178	-0.04988	-0.09988	0.35975
747	0.798	12.54	-0.00	-0.168	-0.167	-0.214	-0.06077	-0.12224	0.35811
747	0.799	0.05	-0.00	-0.104	-0.140	-0.141	-0.03757	-0.08999	0.32944
748	0.800	-3.08	-0.00	-0.109	-0.114	-0.127	-0.03933	-0.07745	0.28830

TABLE VA
BALANCE CAVITY AND BASE PRESSURE COEFFICIENTS, AND INCREMENTAL AXIAL FORCE COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	M_c	α	ϕ	C_{p_c}	$C_{p_{b1}}$	$C_{p_{b2}}$	ΔC_{A_c}	ΔC_{A_b}	C_{A_u}
748	6	0.799	-0.05	-0.00	-0.104	-0.122	-0.125	-0.03779	-0.07953	0.28328
748	7	0.798	0.01	-0.00	-0.108	-0.125	-0.125	-0.03898	-0.08051	0.28393
748	8	0.79A	0.97	-0.00	-0.109	-0.129	-0.127	-0.03959	-0.08233	0.28730
74A	9	0.801	2.06	-0.00	-0.108	-0.126	-0.129	-0.03908	-0.08198	0.29061
74A	10	0.801	3.11	-0.00	-0.108	-0.126	-0.137	-0.03915	-0.08435	0.29061
74A	11	0.799	4.16	-0.00	-0.114	-0.129	-0.141	-0.04108	-0.08666	0.29023
74A	12	0.800	-0.02	-0.00	-0.105	-0.124	-0.123	-0.03793	-0.07916	0.28252
749	1	0.799	-3.10	-0.00	-0.109	-0.110	-0.125	-0.03954	-0.07543	0.28617
749	2	0.800	0.00	-0.00	-0.106	-0.125	-0.123	-0.03843	-0.07964	0.28239
749	3	0.800	0.49	-0.00	-0.105	-0.122	-0.117	-0.03798	-0.07678	0.28361
749	4	0.798	1.03	-0.00	-0.108	-0.126	-0.119	-0.03913	-0.07901	0.28538
749	5	0.800	2.05	-0.00	-0.111	-0.128	-0.123	-0.04019	-0.08061	0.28827
749	6	0.79A	3.10	-0.00	-0.116	-0.129	-0.125	-0.04176	-0.08152	0.28960
749	7	0.798	4.15	-0.00	-0.114	-0.125	-0.129	-0.04103	-0.08149	0.28935
749	8	0.799	-0.06	-0.00	-0.107	-0.124	-0.123	-0.03869	-0.07946	0.28293
750	1	0.798	-3.10	-0.00	-0.113	-0.114	-0.129	-0.04069	-0.07815	0.28273
750	2	0.799	-0.05	-0.00	-0.105	-0.124	-0.122	-0.03808	-0.07895	0.28160
750	3	0.798	0.49	-0.00	-0.106	-0.127	-0.121	-0.03828	-0.07901	0.28164
750	4	0.79A	1.00	-0.00	-0.112	-0.128	-0.123	-0.03960	-0.08036	0.28437
750	5	0.799	2.05	-0.00	-0.108	-0.120	-0.126	-0.04035	-0.08141	0.28672
750	6	0.799	3.11	-0.00	-0.109	-0.120	-0.127	-0.03916	-0.07937	0.28584
750	7	0.799	4.15	-0.00	-0.107	-0.118	-0.133	-0.03933	-0.08053	0.28575
750	8	0.798	-0.02	-0.00	-0.107	-0.124	-0.123	-0.03873	-0.08006	0.28131
751	1	0.797	-3.10	-0.00	-0.111	-0.112	-0.124	-0.04013	-0.07566	0.28296
751	2	0.79A	-0.04	-0.00	-0.10A	-0.123	-0.122	-0.03889	-0.07876	0.28202
751	3	0.797	0.51	-0.00	-0.110	-0.129	-0.125	-0.03986	-0.08182	0.28134
751	4	0.798	1.02	-0.00	-0.106	-0.125	-0.120	-0.03820	-0.07855	0.28319
751	5	0.797	2.05	-0.00	-0.111	-0.127	-0.126	-0.04012	-0.08124	0.28616
751	6	0.799	3.10	-0.00	-0.110	-0.123	-0.125	-0.03974	-0.07982	0.28683
751	7	0.797	4.14	-0.00	-0.112	-0.122	-0.134	-0.04050	-0.08202	0.28633
751	8	0.798	-0.07	-0.00	-0.106	-0.124	-0.124	-0.03850	-0.07992	0.27969
752	1	0.798	-3.10	-0.00	-0.112	-0.114	-0.126	-0.04062	-0.07710	0.28230
752	2	0.797	-0.06	-0.00	-0.110	-0.127	-0.123	-0.03990	-0.08168	0.28044
752	3	0.79A	0.49	-0.00	-0.109	-0.127	-0.123	-0.03930	-0.08046	0.28115
752	4	0.79A	1.02	-0.00	-0.111	-0.129	-0.125	-0.04022	-0.08154	0.28523
752	5	0.79A	2.05	-0.00	-0.111	-0.125	-0.129	-0.04002	-0.08170	0.28578
752	6	0.799	3.10	-0.00	-0.110	-0.122	-0.132	-0.03987	-0.08142	0.28705
752	7	0.798	4.16	-0.00	-0.114	-0.124	-0.140	-0.04131	-0.08485	0.28720

TABLE VA
BALANCE CAVITY AND BASE PRESSURE COEFFICIENTS, AND INCREMENTAL AXIAL FORCE COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	M _c	α	φ	C _{Pc}	C _{Pb1}	C _{Pb2}	ΔC _{Ac}	ΔC _{Ab}	C _{Au}
752	0.799	-0.04	-0.00	-0.106	-0.123	-0.123	-0.03820	-0.07875	0.28084
753	0.799	-3.11	-0.00	-0.112	-0.112	-0.127	-0.04044	-0.07658	0.28399
753	0.799	-0.02	-0.00	-0.108	-0.123	-0.124	-0.03891	-0.07928	0.27995
753	0.79A	0.50	-0.00	-0.110	-0.128	-0.123	-0.03982	-0.08064	0.28131
753	0.800	1.02	-0.00	-0.107	-0.124	-0.120	-0.03887	-0.07850	0.28208
753	0.79A	2.06	-0.00	-0.113	-0.129	-0.128	-0.04097	-0.08240	0.28583
753	0.799	3.09	-0.00	-0.111	-0.123	-0.128	-0.04017	-0.08062	0.28787
753	0.799	4.15	-0.00	-0.112	-0.121	-0.134	-0.04041	-0.08180	0.28680
753	0.800	-0.04	-0.00	-0.104	-0.119	-0.121	-0.03763	-0.07702	0.28019
754	0.800	-3.10	-0.00	-0.110	-0.116	-0.124	-0.03961	-0.07703	0.28384
754	0.799	-0.05	-0.00	-0.107	-0.123	-0.125	-0.03882	-0.07970	0.28089
754	0.800	0.42	-0.00	-0.107	-0.124	-0.123	-0.03860	-0.07932	0.28225
754	0.799	1.06	-0.00	-0.109	-0.127	-0.129	-0.03948	-0.08234	0.28501
754	0.800	2.06	-0.00	-0.114	-0.130	-0.140	-0.04121	-0.08657	0.28680
754	0.800	3.10	-0.00	-0.110	-0.122	-0.137	-0.03983	-0.08305	0.28681
754	0.800	4.13	-0.00	-0.113	-0.121	-0.142	-0.04079	-0.08447	0.28737
754	0.800	-0.03	-0.00	-0.108	-0.124	-0.126	-0.03910	-0.08020	0.28125
755	0.800	-3.10	-0.00	-0.109	-0.110	-0.123	-0.03929	-0.07495	0.28462
755	0.800	-0.04	-0.00	-0.106	-0.123	-0.124	-0.03840	-0.07921	0.28128
755	0.801	0.42	-0.00	-0.106	-0.127	-0.126	-0.03982	-0.08148	0.28210
755	0.801	1.05	-0.00	-0.109	-0.125	-0.122	-0.03950	-0.07912	0.28427
755	0.800	2.10	-0.00	-0.113	-0.127	-0.126	-0.03959	-0.08063	0.28826
755	0.799	3.14	-0.00	-0.116	-0.127	-0.133	-0.04077	-0.08344	0.28999
755	0.801	-0.01	-0.00	-0.104	-0.126	-0.138	-0.04176	-0.08476	0.28999
756	0.799	-3.10	-0.00	-0.111	-0.121	-0.120	-0.03773	-0.07758	0.28053
756	0.79A	0.42	-0.00	-0.104	-0.127	-0.123	-0.04028	-0.08041	0.29212
756	0.800	0.49	-0.00	-0.104	-0.125	-0.129	-0.03879	-0.08163	0.28609
756	0.79A	1.00	-0.00	-0.112	-0.130	-0.145	-0.04039	-0.08230	0.29051
756	0.799	2.05	-0.00	-0.109	-0.128	-0.146	-0.03938	-0.08769	0.29383
756	0.800	3.16	-0.00	-0.110	-0.127	-0.143	-0.03972	-0.08685	0.29222
756	0.799	-0.05	-0.00	-0.106	-0.124	-0.143	-0.03982	-0.08513	0.29172
757	0.799	-3.11	-0.00	-0.116	-0.119	-0.131	-0.04203	-0.08011	0.29684
757	0.800	-0.05	-0.00	-0.106	-0.126	-0.124	-0.03847	-0.08043	0.29216
757	0.800	0.48	-0.00	-0.106	-0.127	-0.121	-0.03824	-0.07976	0.29161
757	0.800	1.02	-0.00	-0.108	-0.131	-0.130	-0.03914	-0.08367	0.29402

TABLE VA
BALANCE CAVITY AND BASE PRESSURE COEFFICIENTS, AND INCREMENTAL
AXIAL FORCE COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	M _c	α	φ	C _p c	C _p b _l	C _p b ₂	ΔC _A c	ΔC _A b	C _A u
757	5	0.799	2.05	-0.00	-0.112	-0.131	-0.133	-0.04056	-0.08475	0.29813
757	6	0.79A	3.09	0.00	-0.119	-0.137	-0.130	-0.04317	-0.08571	0.29991
757	7	0.800	4.14	-0.00	-0.114	-0.130	-0.127	-0.04122	-0.08276	0.29680
757	8	0.800	-0.01	-0.00	-0.104	-0.123	-0.120	-0.03746	-0.07802	0.28999
75A	3	0.600	-3.12	-0.00	-0.092	-0.101	-0.097	-0.03318	-0.06379	0.28377
75A	4	0.601	-1.07	-0.00	-0.085	-0.103	-0.101	-0.03085	-0.06553	0.27950
75A	5	0.601	-0.06	-0.00	-0.085	-0.107	-0.100	-0.03062	-0.06560	0.27939
75A	6	0.600	0.49	-0.00	-0.086	-0.112	-0.101	-0.03127	-0.06844	0.27955
75A	7	0.600	0.97	-0.00	-0.089	-0.114	-0.098	-0.03213	-0.06807	0.27908
75A	8	0.600	3.09	-0.00	-0.089	-0.110	-0.100	-0.03234	-0.06764	0.27754
75A	9	0.600	6.19	-0.00	-0.095	-0.108	-0.104	-0.03423	-0.07462	0.28603
75A	10	0.600	9.27	-0.00	-0.114	-0.110	-0.103	-0.04129	-0.08707	0.29706
75A	11	0.600	12.44	-0.00	-0.145	-0.149	-0.178	-0.05222	-0.10484	0.29706
75A	12	0.601	-0.06	-0.00	-0.085	-0.108	-0.100	-0.03062	-0.06694	0.27815
759	1	0.600	-3.09	-0.00	-0.090	-0.100	-0.100	-0.03268	-0.06419	0.27862
759	2	0.600	-1.09	-0.00	-0.088	-0.105	-0.104	-0.03182	-0.06700	0.27948
759	3	0.600	-0.05	-0.00	-0.088	-0.109	-0.102	-0.03184	-0.06787	0.27891
759	4	0.600	0.49	-0.00	-0.086	-0.115	-0.099	-0.03115	-0.06696	0.27802
759	5	0.600	0.98	-0.00	-0.086	-0.115	-0.099	-0.03183	-0.06869	0.27915
759	6	0.600	3.11	-0.00	-0.090	-0.109	-0.104	-0.03266	-0.06867	0.27855
759	7	0.601	6.18	-0.00	-0.094	-0.105	-0.122	-0.03404	-0.07299	0.28119
759	8	0.600	9.27	-0.00	-0.114	-0.118	-0.150	-0.04114	-0.08602	0.28954
759	9	0.599	12.46	-0.00	-0.148	-0.151	-0.184	-0.05331	-0.10730	0.30454
759	10	0.599	-0.03	-0.00	-0.086	-0.109	-0.100	-0.03130	-0.06723	0.27684
760	1	0.600	-3.08	-0.00	-0.087	-0.096	-0.096	-0.03156	-0.06188	0.27626
760	2	0.600	-1.07	-0.00	-0.085	-0.100	-0.099	-0.03085	-0.06398	0.27902
760	3	0.600	-0.03	-0.00	-0.087	-0.108	-0.096	-0.03141	-0.06550	0.27815
760	4	0.600	0.48	-0.00	-0.085	-0.105	-0.094	-0.03089	-0.06407	0.27826
760	5	0.599	0.99	-0.00	-0.088	-0.111	-0.096	-0.03189	-0.06673	0.28180
760	6	0.600	3.11	-0.00	-0.090	-0.108	-0.104	-0.03269	-0.06810	0.28070
760	7	0.600	6.19	-0.00	-0.097	-0.106	-0.125	-0.03501	-0.07426	0.28361
760	8	0.599	9.29	-0.00	-0.115	-0.116	-0.153	-0.04135	-0.08617	0.29301
760	9	0.599	12.44	-0.00	-0.145	-0.150	-0.180	-0.05235	-0.10591	0.30709
760	10	0.599	-0.03	-0.00	-0.088	-0.109	-0.096	-0.03189	-0.06606	0.27836
761	1	0.600	-3.07	-0.00	-0.088	-0.095	-0.094	-0.03179	-0.06070	0.27589
761	2	0.599	-1.05	-0.00	-0.084	-0.098	-0.097	-0.03025	-0.06291	0.27942
761	3	0.599	-0.04	-0.00	-0.084	-0.109	-0.099	-0.03174	-0.06676	0.27892
761	4	0.600	0.49	-0.00	-0.085	-0.106	-0.098	-0.03081	-0.06556	0.27924

TABLE VA
BALANCE CAVITY AND BASE PRESSURE COEFFICIENTS, AND INCREMENTAL
AXIAL FORCE COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	M _C	α	φ	C _{Pc}	C _{Pb1}	C _{Pb2}	ΔC _{Ac}	ΔC _{Ab}	C _{Au}
761	5	0.599	-0.00	-0.090	-0.108	-0.098	-0.03240	-0.06618	0.28148
761	6	0.599	-0.00	-0.090	-0.109	-0.103	-0.03274	-0.06606	0.28376
761	7	0.599A	-0.00	-0.100	-0.108	-0.126	-0.03617	-0.07523	0.28675
761	8	0.600	-0.00	-0.114	-0.118	-0.153	-0.04127	-0.08675	0.29693
761	9	0.600	-0.00	-0.146	-0.149	-0.177	-0.05257	-0.10454	0.31033
761	10	0.602	-0.00	-0.086	-0.105	-0.096	-0.03110	-0.06466	0.27735
762	1	0.601	-0.00	-0.090	-0.099	-0.101	-0.03274	-0.06433	0.27533
762	2	0.601	-0.00	-0.085	-0.101	-0.102	-0.03091	-0.06524	0.28127
762	3	0.602	-0.00	-0.087	-0.106	-0.095	-0.03071	-0.06461	0.27672
762	4	0.601	-0.00	-0.087	-0.109	-0.098	-0.03145	-0.06657	0.28201
762	5	0.601	-0.00	-0.086	-0.106	-0.092	-0.03127	-0.06386	0.28332
762	6	0.601	-0.00	-0.090	-0.103	-0.102	-0.03271	-0.06592	0.28910
762	7	0.600	-0.00	-0.101	-0.109	-0.127	-0.03641	-0.07583	0.29405
762	8	0.601	-0.00	-0.114	-0.116	-0.149	-0.04130	-0.08521	0.30094
762	9	0.601	-0.00	-0.141	-0.142	-0.176	-0.05089	-0.10199	0.31485
762	10	0.601	-0.00	-0.087	-0.107	-0.098	-0.03135	-0.06575	0.27746
763	1	0.601	-0.00	-0.089	-0.098	-0.100	-0.03219	-0.06383	0.27658
763	2	0.601	-0.00	-0.084	-0.104	-0.104	-0.03025	-0.06663	0.28611
763	3	0.601	-0.00	-0.083	-0.107	-0.099	-0.03008	-0.06619	0.28594
763	4	0.601	-0.00	-0.084	-0.108	-0.092	-0.03025	-0.06412	0.28672
763	5	0.601	-0.00	-0.086	-0.111	-0.096	-0.03116	-0.06640	0.29063
763	6	0.601	-0.00	-0.095	-0.108	-0.108	-0.03443	-0.06930	0.30053
763	7	0.601	-0.00	-0.098	-0.103	-0.120	-0.03539	-0.07194	0.30492
763	8	0.600	-0.00	-0.110	-0.113	-0.148	-0.03995	-0.08390	0.31392
763	9	0.600	-0.00	-0.141	-0.146	-0.178	-0.05092	-0.10383	0.32315
763	10	0.602	-0.00	-0.084	-0.108	-0.100	-0.03031	-0.06691	0.28451
764	1	0.602	-0.00	-0.087	-0.099	-0.106	-0.03162	-0.06585	0.28336
764	2	0.602	-0.00	-0.085	-0.119	-0.111	-0.03174	-0.07113	0.29542
764	3	0.601	-0.00	-0.087	-0.119	-0.108	-0.03151	-0.07313	0.29802
764	4	0.601	-0.00	-0.086	-0.116	-0.100	-0.03123	-0.06959	0.29777
764	5	0.601	-0.00	-0.091	-0.117	-0.099	-0.03280	-0.05969	0.29941
764	6	0.602	-0.00	-0.094	-0.108	-0.121	-0.03402	-0.07050	0.31223
764	7	0.602	-0.00	-0.097	-0.103	-0.111	-0.03507	-0.07216	0.31543
764	8	0.602	-0.00	-0.110	-0.113	-0.148	-0.03988	-0.08379	0.31968
764	9	0.601	-0.00	-0.138	-0.141	-0.177	-0.04990	-0.10198	0.32779
764	10	0.601	-0.00	-0.086	-0.115	-0.108	-0.03101	-0.07173	0.29631
765	1	0.602	-0.00	-0.084	-0.098	-0.103	-0.03051	-0.06484	0.29407
765	2	0.601	-0.00	-0.084	-0.113	-0.116	-0.03052	-0.07378	0.30711

TABLE VA
 BALANCE CAVITY AND BASE PRESSURE COEFFICIENTS, AND INCREMENTAL
 AXIAL FORCE COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	M _c	α	Φ	C _{p,c}	C _{p,b1}	C _{p,b2}	ΔC _{A,c}	ΔC _{A,b}	C _{A,u}
765	0.601	0.02	-0.00	-0.086	-0.121	-0.109	-0.03127	-0.07397	0.31070
765	0.600	0.55	-0.00	-0.084	-0.120	-0.104	-0.03047	-0.07176	0.31205
765	0.601	1.05	-0.00	-0.085	-0.116	-0.099	-0.03074	-0.06887	0.31313
765	0.600	3.17	-0.00	-0.096	-0.111	-0.114	-0.03484	-0.07247	0.32523
765	0.600	6.24	-0.00	-0.099	-0.106	-0.124	-0.03576	-0.07375	0.32786
765	0.600	9.31	-0.00	-0.110	-0.112	-0.147	-0.03977	-0.08344	0.32883
765	0.600	12.46	-0.00	-0.136	-0.139	-0.174	-0.04911	-0.10027	0.35518
765	0.601	10.00	-0.00	-0.085	-0.120	-0.111	-0.03092	-0.07430	0.31237
766	0.600	3.00	-0.00	-0.088	-0.101	-0.111	-0.03168	-0.06789	0.30611
766	0.601	-0.96	-0.00	-0.087	-0.115	-0.122	-0.03132	-0.07620	0.32182
766	0.601	0.55	-0.00	-0.090	-0.127	-0.118	-0.03246	-0.07882	0.32584
766	0.601	1.05	-0.00	-0.086	-0.127	-0.109	-0.03104	-0.07582	0.32735
766	0.600	3.18	-0.00	-0.089	-0.126	-0.104	-0.03205	-0.07399	0.32693
766	0.600	6.24	-0.00	-0.096	-0.112	-0.117	-0.03487	-0.07366	0.33305
766	0.600	9.31	-0.00	-0.100	-0.105	-0.123	-0.03613	-0.07327	0.33881
766	0.601	12.46	-0.00	-0.112	-0.116	-0.149	-0.04062	-0.08514	0.34271
766	0.601	10.00	-0.00	-0.134	-0.138	-0.174	-0.04842	-0.10012	0.34297
767	0.600	3.07	-0.00	-0.087	-0.125	-0.117	-0.03165	-0.07770	0.30055
767	0.601	-0.05	-0.00	-0.083	-0.094	-0.093	-0.03015	-0.06024	0.27909
767	0.601	0.45	-0.00	-0.082	-0.107	-0.093	-0.02970	-0.06428	0.28145
767	0.602	0.99	-0.00	-0.085	-0.109	-0.096	-0.03060	-0.06333	0.28402
767	0.600	2.04	-0.00	-0.082	-0.108	-0.101	-0.02962	-0.06727	0.28591
767	0.600	3.08	-0.00	-0.087	-0.112	-0.107	-0.03156	-0.07032	0.28591
767	0.600	4.11	-0.00	-0.091	-0.113	-0.108	-0.03262	-0.07094	0.28047
767	0.600	-0.07	-0.00	-0.085	-0.103	-0.095	-0.03071	-0.06351	0.27831
768	0.601	3.09	-0.00	-0.084	-0.093	-0.092	-0.03026	-0.05946	0.27874
768	0.600	-0.46	-0.00	-0.086	-0.108	-0.094	-0.03105	-0.06478	0.28187
768	0.600	1.04	-0.00	-0.085	-0.109	-0.091	-0.03072	-0.06421	0.28164
768	0.600	2.04	-0.00	-0.089	-0.111	-0.090	-0.03215	-0.06390	0.28157
768	0.601	3.08	-0.00	-0.085	-0.105	-0.090	-0.03080	-0.06256	0.28399
768	0.600	4.12	-0.00	-0.090	-0.106	-0.098	-0.03263	-0.06551	0.27748
768	0.603	-0.103	-0.00	-0.080	-0.101	-0.090	-0.02891	-0.06173	0.27702
769	0.601	3.09	-0.00	-0.087	-0.098	-0.094	-0.03157	-0.06167	0.27812
769	0.601	-0.06	-0.00	-0.084	-0.106	-0.093	-0.03040	-0.06392	0.28134
769	0.601	0.45	-0.00	-0.085	-0.107	-0.091	-0.03078	-0.06366	0.28047
769	0.601	1.00	-0.00	-0.085	-0.107	-0.094	-0.03083	-0.06454	0.28047

TABLE VA
 BALANCE CAVITY AND BASE PRESSURE COEFFICIENTS, AND INCREMENTAL
 AXIAL FORCE COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	M _c	α	Φ	C _{Pc}	C _{Pb1}	C _{Pb2}	ΔC _{Ac}	ΔC _{Ab}	C _{Au}
769	0.601	2.05	-0.00	-0.091	-0.109	-0.090	-0.03204	-0.06669	0.20101
769	0.601	3.10	-0.00	-0.088	-0.104	-0.099	-0.03170	-0.06520	0.20173
769	0.600	4.11	-0.00	-0.089	-0.104	-0.105	-0.03230	-0.06745	0.20260
769	0.600	-0.01	-0.00	-0.087	-0.105	-0.098	-0.03132	-0.06539	0.27606
770	0.600	-3.08	-0.00	-0.085	-0.096	-0.091	-0.03060	-0.06025	0.27714
770	0.599	-0.05	-0.00	-0.087	-0.105	-0.095	-0.03148	-0.06425	0.27029
770	0.600	0.44	-0.00	-0.089	-0.110	-0.096	-0.03205	-0.06626	0.20131
770	0.600	1.03	-0.00	-0.085	-0.108	-0.092	-0.03074	-0.06422	0.20022
770	0.600	3.05	-0.00	-0.092	-0.111	-0.101	-0.03329	-0.06030	0.20203
770	0.600	3.10	-0.00	-0.088	-0.106	-0.090	-0.03101	-0.06564	0.20230

TABLE VA
BALANCE CAVITY AND BASE PRESSURE COEFFICIENTS, AND INCREMENTAL AXIAL FORCE COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	M _c	α	Φ	C _{p,c}	C _{p,b1}	C _{p,b2}	ΔC _{A,c}	ΔC _{A,b}	C _{A,u}
770	0.600	4.14	-0.00	-0.090	-0.108	-0.103	-0.03270	-0.06771	0.28152
770	0.601	-0.02	-0.00	-0.082	-0.103	-0.093	-0.02965	-0.06305	0.27736
771	0.600	-3.09	-0.00	-0.087	-0.098	-0.092	-0.03166	-0.06120	0.27861
771	0.601	-1.07	-0.00	-0.085	-0.102	-0.095	-0.03073	-0.06322	0.27953
771	0.600	-0.03	-0.00	-0.082	-0.102	-0.093	-0.02983	-0.06274	0.27820
771	0.601	0.46	-0.00	-0.085	-0.106	-0.095	-0.03095	-0.06411	0.28062
771	0.600	0.9A	-0.00	-0.086	-0.109	-0.095	-0.03125	-0.06563	0.28088
771	0.599	2.04	-0.00	-0.08A	-0.108	-0.098	-0.03190	-0.06627	0.28158
771	0.600	3.10	-0.00	-0.087	-0.106	-0.104	-0.03152	-0.06750	0.28235
771	0.600	4.11	-0.00	-0.089	-0.105	-0.106	-0.03226	-0.06786	0.28341
771	0.601	-0.07	-0.00	-0.084	-0.103	-0.095	-0.03041	-0.06378	0.27911
772	0.600	-3.08	-0.00	-0.087	-0.100	-0.093	-0.03145	-0.06194	0.27649
772	0.602	-0.08	-0.00	-0.084	-0.105	-0.092	-0.03037	-0.06340	0.27990
772	0.601	0.46	-0.00	-0.084	-0.106	-0.093	-0.03044	-0.06417	0.28130
772	0.601	0.99	-0.00	-0.085	-0.108	-0.093	-0.03089	-0.06434	0.28011
772	0.600	2.03	-0.00	-0.087	-0.107	-0.096	-0.03162	-0.06533	0.28184
772	0.602	3.10	-0.00	-0.087	-0.107	-0.096	-0.0310A	-0.06418	0.28141
772	0.601	4.14	-0.00	-0.086	-0.107	-0.104	-0.03145	-0.06785	0.28231
772	0.601	-0.03	-0.00	-0.084	-0.105	-0.094	-0.03034	-0.06430	0.27798
773	0.601	-3.10	-0.00	-0.086	-0.099	-0.095	-0.03111	-0.06230	0.27857
773	0.601	-0.04	-0.00	-0.082	-0.105	-0.094	-0.02981	-0.06415	0.27933
773	0.602	0.46	-0.00	-0.082	-0.102	-0.093	-0.02960	-0.06280	0.28164
773	0.601	0.99	-0.00	-0.084	-0.107	-0.096	-0.03032	-0.06523	0.28050
773	0.602	2.03	-0.00	-0.085	-0.105	-0.096	-0.03065	-0.06469	0.28208
773	0.601	3.08	-0.00	-0.085	-0.106	-0.107	-0.03099	-0.06829	0.28392
773	0.601	4.12	-0.00	-0.086	-0.110	-0.109	-0.03195	-0.07018	0.28592
773	0.601	-0.01	-0.00	-0.083	-0.104	-0.095	-0.02992	-0.06384	0.27742
774	0.601	-3.10	-0.00	-0.086	-0.100	-0.093	-0.03125	-0.06208	0.27868
774	0.601	-0.04	-0.00	-0.085	-0.108	-0.095	-0.03064	-0.06547	0.27893
774	0.602	0.45	-0.00	-0.084	-0.106	-0.094	-0.03036	-0.06401	0.28049
774	0.601	1.02	-0.00	-0.084	-0.108	-0.091	-0.03059	-0.06413	0.28227
774	0.601	2.03	-0.00	-0.086	-0.109	-0.098	-0.03078	-0.06636	0.28322
774	0.602	3.10	-0.00	-0.085	-0.107	-0.098	-0.03178	-0.06572	0.28408
774	0.601	4.12	-0.00	-0.08A	-0.108	-0.108	-0.03183	-0.06917	0.28425
774	0.601	-0.03	-0.00	-0.082	-0.107	-0.095	-0.02982	-0.06479	0.27966
775	0.601	-3.10	-0.00	-0.087	-0.108	-0.095	-0.03152	-0.06531	0.28489
775	0.601	-0.04	-0.00	-0.080	-0.104	-0.097	-0.02901	-0.06461	0.28353

TABLE VA
 BALANCE CAVITY AND BASE PRESSURE COEFFICIENTS, AND INCREMENTAL
 AXIAL FORCE COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	M_c	α	Φ	C_{p_c}	$C_{p_{b1}}$	$C_{p_{b2}}$	ΔC_{A_c}	ΔC_{A_b}	C_{A_u}
775	3	0.601	0.46	-0.00	-0.084	-0.110	-0.105	-0.03057	-0.06916	0.28639
775	4	0.601	1.03	-0.00	-0.085	-0.108	-0.107	-0.02977	-0.06913	0.28594
775	5	0.601	3.08	0.00	-0.087	-0.111	-0.116	-0.03085	-0.07280	0.28857
775	6	0.601	4.12	-0.00	-0.087	-0.111	-0.118	-0.03153	-0.07355	0.28971
775	7	0.601	-0.05	-0.00	-0.085	-0.107	-0.102	-0.03067	-0.06743	0.28372
776	1	0.600	3.10	-0.00	-0.088	-0.100	-0.097	-0.03197	-0.06318	0.29005
776	2	0.601	-0.03	-0.00	-0.082	-0.106	-0.094	-0.02959	-0.06422	0.28515
776	3	0.600	1.02	-0.00	-0.086	-0.113	-0.096	-0.03099	-0.06709	0.28937
776	4	0.601	2.02	-0.00	-0.088	-0.117	-0.095	-0.03110	-0.06664	0.28963
776	5	0.601	3.09	-0.00	-0.091	-0.122	-0.101	-0.03185	-0.07008	0.29205
776	6	0.601	4.11	-0.00	-0.089	-0.118	-0.100	-0.03300	-0.07153	0.29472
776	7	0.601	-0.05	-0.00	-0.082	-0.109	-0.098	-0.03218	-0.06524	0.28656
776	8	1.251	3.17	-0.00	-0.210	-0.214	-0.224	-0.07563	-0.13913	0.53409
776	9	1.251	1.05	-0.00	-0.204	-0.217	-0.219	-0.07355	-0.13888	0.52249
776	10	1.251	3.12	-0.00	-0.205	-0.217	-0.222	-0.07385	-0.14069	0.52210
777	3	1.251	6.26	-0.00	-0.208	-0.218	-0.227	-0.07520	-0.14256	0.52306
777	4	1.251	9.39	-0.00	-0.251	-0.236	-0.245	-0.08199	-0.14853	0.54170
777	5	1.251	12.67	-0.00	-0.273	-0.266	-0.268	-0.09047	-0.16156	0.56012
777	6	1.251	-0.07	-0.00	-0.209	-0.220	-0.226	-0.07547	-0.14306	0.52142
779	1	1.251	3.16	-0.00	-0.215	-0.215	-0.229	-0.07755	-0.14244	0.52722
779	2	1.251	-0.05	-0.00	-0.209	-0.219	-0.227	-0.07524	-0.14286	0.51931
779	3	1.251	1.00	-0.00	-0.209	-0.220	-0.228	-0.07546	-0.14367	0.52254
779	4	1.251	3.12	-0.00	-0.231	-0.222	-0.234	-0.07665	-0.14582	0.54352
779	5	1.251	6.24	-0.00	-0.255	-0.238	-0.249	-0.08337	-0.15109	0.56973
779	6	1.251	9.41	-0.00	-0.278	-0.270	-0.271	-0.09190	-0.16158	0.59543
779	7	1.251	12.63	-0.00	-0.207	-0.216	-0.225	-0.07475	-0.14148	0.51176
779	8	1.251	-0.03	-0.00	-0.213	-0.214	-0.229	-0.07696	-0.14192	0.52351
780	1	1.251	3.13	-0.00	-0.208	-0.218	-0.225	-0.07535	-0.14192	0.52104
780	2	1.250	-0.02	-0.00	-0.209	-0.220	-0.227	-0.07535	-0.14332	0.52510
780	3	1.251	1.04	-0.00	-0.214	-0.220	-0.234	-0.07722	-0.14575	0.54561
780	4	1.251	3.16	-0.00	-0.232	-0.222	-0.249	-0.08364	-0.15075	0.57157
780	5	1.250	6.22	-0.00	-0.255	-0.238	-0.271	-0.09193	-0.16315	0.59837
780	6	1.250	9.44	-0.00	-0.278	-0.270	-0.297	-0.10041	-0.18154	0.62154
780	7	1.251	12.64	-0.00	-0.278	-0.270	-0.297	-0.10041	-0.18154	0.62154

TABLE VA
 BALANCE CAVITY AND BASE PRESSURE COEFFICIENTS, AND INCREMENTAL
 AXIAL FORCE COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	M _c	α	Φ	C _{p_c}	C _{p_{b1}}	C _{p_{b2}}	ΔC _{A_c}	ΔC _{A_b}	C _{A_u}
780	1.250	-0.02	-0.00	-0.208	-0.218	-0.226	-0.07517	-0.14233	0.51661
781	1.251	-3.11	-0.00	-0.213	-0.213	-0.228	-0.07669	-0.14139	0.52140
781	1.251	-0.04	-0.00	-0.210	-0.219	-0.226	-0.07579	-0.14278	0.51847
781	1.251	1.03	-0.00	-0.211	-0.221	-0.229	-0.07606	-0.14430	0.52250
781	1.251	3.17	-0.00	-0.215	-0.220	-0.235	-0.07748	-0.14623	0.52878
781	1.249	6.30	-0.00	-0.235	-0.233	-0.251	-0.08464	-0.15226	0.54867
781	1.250	9.43	-0.00	-0.255	-0.237	-0.271	-0.09202	-0.16280	0.57614
781	1.250	12.64	-0.00	-0.280	-0.270	-0.298	-0.10106	-0.18204	0.60355
781	1.251	-0.05	-0.00	-0.210	-0.219	-0.227	-0.07584	-0.14303	0.51805
782	1.250	-3.10	-0.00	-0.215	-0.216	-0.231	-0.07768	-0.14342	0.52106
782	1.250	-0.01	-0.00	-0.215	-0.223	-0.229	-0.07745	-0.14500	0.52491
782	1.250	1.05	-0.00	-0.215	-0.225	-0.232	-0.07767	-0.14653	0.52794
782	1.250	3.18	-0.00	-0.221	-0.226	-0.243	-0.07983	-0.15025	0.53733
782	1.251	6.32	-0.00	-0.233	-0.222	-0.253	-0.08407	-0.15225	0.55566
782	1.250	9.45	-0.00	-0.257	-0.238	-0.273	-0.09275	-0.16382	0.58340
782	1.251	12.66	-0.00	-0.282	-0.270	-0.299	-0.10162	-0.18251	0.61249
782	1.251	-0.03	-0.00	-0.214	-0.223	-0.230	-0.07736	-0.14511	0.52234
783	1.251	-3.08	-0.00	-0.216	-0.222	-0.236	-0.07789	-0.14670	0.52941
783	1.250	0.00	-0.00	-0.220	-0.231	-0.237	-0.07948	-0.14984	0.54292
783	1.249	1.08	-0.00	-0.219	-0.232	-0.238	-0.07905	-0.15073	0.54411
783	1.249	3.19	-0.00	-0.227	-0.234	-0.250	-0.08174	-0.15512	0.55480
783	1.251	6.33	-0.00	-0.231	-0.229	-0.255	-0.08340	-0.15355	0.57383
783	1.251	9.47	-0.00	-0.254	-0.239	-0.274	-0.09153	-0.16436	0.60036
783	1.250	12.64	-0.00	-0.282	-0.272	-0.301	-0.10168	-0.18390	0.63133
783	1.251	0.00	-0.00	-0.220	-0.232	-0.238	-0.07943	-0.15047	0.54130
784	1.251	-3.07	-0.00	-0.221	-0.228	-0.243	-0.07989	-0.15074	0.54011
784	1.252	0.04	-0.00	-0.224	-0.236	-0.242	-0.08096	-0.15314	0.55662
784	1.251	1.08	-0.00	-0.227	-0.237	-0.245	-0.08099	-0.15388	0.55895
784	1.252	3.22	-0.00	-0.233	-0.234	-0.251	-0.08198	-0.15572	0.56953
784	1.251	6.35	-0.00	-0.252	-0.227	-0.258	-0.08590	-0.15534	0.58809
784	1.252	9.46	-0.00	-0.252	-0.238	-0.274	-0.09088	-0.16424	0.61393
784	1.251	12.69	-0.00	-0.280	-0.271	-0.300	-0.10101	-0.18313	0.64665
784	1.251	-0.03	-0.00	-0.225	-0.237	-0.242	-0.08115	-0.15385	0.55553
785	1.250	-3.01	-0.00	-0.225	-0.232	-0.244	-0.08121	-0.15275	0.56840
785	1.248	0.07	-0.00	-0.232	-0.243	-0.248	-0.08363	-0.15748	0.58920
785	1.250	1.13	-0.00	-0.230	-0.243	-0.248	-0.08307	-0.15726	0.59724
785	1.255	3.22	-0.00	-0.226	-0.235	-0.252	-0.08157	-0.15598	0.60031

TABLE VA
BALANCE CAVITY AND BASE PRESSURE COEFFICIENTS, AND INCREMENTAL AXIAL FORCE COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	M _c	α	Φ	C _{Pc}	C _{Pb1}	C _{Pb2}	ΔC _{Ac}	ΔC _{Ab}	C _{Au}
785	5	1.254	6.37	-0.00	-0.233	-0.226	-0.258	-0.08399	-0.15513	0.61917
785	6	1.249	9.50	-0.00	-0.253	-0.240	-0.276	-0.09140	-0.16554	0.64497
785	7	1.250	12.70	-0.00	-0.280	-0.270	-0.301	-0.10083	-0.18330	0.67587
785	8	1.251	0.06	-0.00	-0.228	-0.239	-0.244	-0.08217	-0.15504	0.58756
786	1	0.899	-3.01	-0.00	-0.114	-0.121	-0.146	-0.04119	-0.08584	0.32389
786	2	0.900	0.03	-0.00	-0.113	-0.142	-0.146	-0.04071	-0.09239	0.34594
786	3	0.900	1.09	-0.00	-0.120	-0.146	-0.140	-0.04328	-0.09192	0.34907
786	4	0.900	3.15	-0.00	-0.127	-0.139	-0.151	-0.04604	-0.09308	0.35523
786	5	0.900	6.25	-0.00	-0.130	-0.128	-0.161	-0.04686	-0.09288	0.35188
786	6	0.899	9.34	-0.00	-0.147	-0.139	-0.189	-0.05316	-0.10535	0.35870
786	7	0.901	12.48	-0.00	-0.180	-0.170	-0.227	-0.06508	-0.12744	0.38922
786	8	0.897	0.03	-0.00	-0.119	-0.147	-0.153	-0.04298	-0.09632	0.34414
787	1	0.900	-3.03	-0.00	-0.113	-0.117	-0.139	-0.04087	-0.08215	0.29881
787	2	0.901	0.00	-0.00	-0.114	-0.140	-0.142	-0.04128	-0.09059	0.32029
787	3	0.899	1.07	-0.00	-0.125	-0.144	-0.140	-0.04514	-0.09134	0.31980
787	4	0.899	3.14	-0.00	-0.129	-0.139	-0.151	-0.04677	-0.09295	0.33181
787	5	0.902	6.24	-0.00	-0.127	-0.126	-0.159	-0.04579	-0.09147	0.33757
787	6	0.900	9.32	-0.00	-0.147	-0.140	-0.189	-0.05297	-0.10553	0.33634
787	7	0.901	12.47	-0.00	-0.162	-0.169	-0.224	-0.06581	-0.12612	0.37173
787	8	0.89A	0.03	-0.00	-0.119	-0.145	-0.144	-0.04315	-0.09271	0.31693
78A	1	0.901	-3.06	-0.00	-0.113	-0.114	-0.134	-0.04079	-0.07953	0.29014
78A	2	0.899	0.00	-0.00	-0.117	-0.139	-0.139	-0.04243	-0.08926	0.30433
78A	3	0.900	3.11	-0.00	-0.127	-0.133	-0.148	-0.04577	-0.09026	0.32297
78A	4	0.899	6.23	-0.00	-0.130	-0.130	-0.164	-0.04683	-0.09438	0.32260
78A	5	0.89A	9.32	-0.00	-0.148	-0.139	-0.190	-0.05344	-0.10564	0.32553
78A	6	0.899	12.46	-0.00	-0.183	-0.171	-0.227	-0.06623	-0.12755	0.36169
78A	7	0.901	-0.00	-0.00	-0.112	-0.134	-0.133	-0.04048	-0.08575	0.30510
789	1	0.900	-3.07	-0.00	-0.115	-0.113	-0.135	-0.04170	-0.07953	0.28849
789	2	0.900	0.03	-0.00	-0.113	-0.129	-0.132	-0.04332	-0.08384	0.29960
789	3	0.899	1.01	-0.00	-0.120	-0.136	-0.138	-0.04332	-0.08782	0.29753
789	4	0.900	3.17	-0.00	-0.122	-0.126	-0.140	-0.04393	-0.08552	0.30954
789	5	0.899	6.22	-0.00	-0.131	-0.132	-0.164	-0.04731	-0.09504	0.31449
789	6	0.899	9.28	-0.00	-0.145	-0.139	-0.188	-0.05236	-0.10474	0.31705
789	7	0.899	12.46	-0.00	-0.189	-0.178	-0.232	-0.06805	-0.13135	0.35380
789	8	0.900	-0.03	-0.00	-0.113	-0.127	-0.131	-0.04092	-0.08316	0.29550
7790	1	0.899	-3.08	-0.00	-0.118	-0.116	-0.137	-0.04256	-0.08107	0.29133
7790	2	0.899	-0.04	-0.00	-0.115	-0.126	-0.135	-0.04163	-0.08423	0.29213

TABLE VA
 BALANCE CAVITY AND BASE PRESSURE COEFFICIENTS, AND INCREMENTAL
 AXIAL FORCE COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	M _c	α	φ	C _{pC}	C _{pB1}	C _{pB2}	ΔC _{A C}	ΔC _{A B}	C _{A u}
790	3	0.900	0.9A	-0.00	-0.118	-0.131	-0.133	-0.04254	-0.08500	0.29479
790	4	0.900	3.10	-0.00	-0.121	-0.129	-0.138	-0.04387	-0.08576	0.30269
790	5	0.901	6.21	-0.00	-0.128	-0.131	-0.162	-0.04636	-0.09403	0.30672
790	6	0.901	9.31	-0.00	-0.145	-0.139	-0.189	-0.05252	-0.12931	0.31315
790	7	0.899	12.45	-0.00	-0.187	-0.175	-0.228	-0.06741	-0.12931	0.34852
790	8	0.900	-0.03	-0.00	-0.114	-0.126	-0.133	-0.04111	-0.08328	0.29173
791	1	0.900	-3.09	-0.00	-0.118	-0.115	-0.133	-0.04260	-0.07972	0.29389
791	2	0.900	-0.04	-0.00	-0.116	-0.128	-0.134	-0.04141	-0.08434	0.29249
791	3	0.900	3.10	-0.00	-0.117	-0.129	-0.133	-0.04183	-0.08547	0.29608
791	4	0.901	6.19	-0.00	-0.129	-0.131	-0.137	-0.04230	-0.08523	0.30033
791	5	0.901	9.29	-0.00	-0.149	-0.141	-0.163	-0.04675	-0.09456	0.30518
791	6	0.901	12.46	-0.00	-0.191	-0.179	-0.189	-0.05328	-0.10583	0.31037
791	7	0.900	-0.04	-0.00	-0.113	-0.127	-0.231	-0.06903	-0.13166	0.34585
791	8	0.900	-0.00	-0.00	-0.113	-0.127	-0.133	-0.04096	-0.08353	0.29154
792	1	0.902	-3.11	-0.00	-0.119	-0.118	-0.135	-0.04295	-0.08125	0.30535
792	2	0.899	-0.06	-0.00	-0.112	-0.124	-0.133	-0.04053	-0.08272	0.29483
792	3	0.899	3.07	-0.00	-0.116	-0.133	-0.132	-0.04208	-0.08537	0.29501
792	4	0.899	6.15	-0.00	-0.117	-0.131	-0.135	-0.04216	-0.08548	0.29468
792	5	0.898	9.27	-0.00	-0.128	-0.140	-0.161	-0.04627	-0.09433	0.30305
792	6	0.899	12.45	-0.00	-0.186	-0.175	-0.190	-0.05336	-0.10594	0.33776
792	7	0.900	-0.07	-0.00	-0.113	-0.126	-0.228	-0.06700	-0.12905	0.29443
792	8	0.900	-0.00	-0.00	-0.113	-0.126	-0.134	-0.04100	-0.08335	0.28669
793	6	0.599	-3.12	-0.00	-0.089	-0.099	-0.098	-0.03235	-0.06353	0.28669
793	7	0.601	-0.05	-0.00	-0.081	-0.106	-0.097	-0.02949	-0.06503	0.28045
793	8	0.599	1.00	-0.00	-0.082	-0.110	-0.096	-0.02979	-0.06607	0.28147
793	9	0.599	3.06	-0.00	-0.088	-0.107	-0.102	-0.03186	-0.06850	0.28124
793	10	0.598	6.18	-0.00	-0.096	-0.105	-0.126	-0.03481	-0.07481	0.28092
793	11	0.600	9.28	-0.00	-0.112	-0.115	-0.151	-0.04038	-0.08557	0.28703
793	12	0.599	12.45	-0.00	-0.145	-0.150	-0.180	-0.05232	-0.10597	0.29876
793	13	0.599	-0.03	-0.00	-0.083	-0.107	-0.097	-0.03010	-0.06567	0.28230
794	1	0.599	-3.11	-0.00	-0.088	-0.095	-0.099	-0.03177	-0.06235	0.27979
794	2	0.599	-0.01	-0.00	-0.090	-0.112	-0.103	-0.03249	-0.06903	0.28158
794	3	0.598	1.02	-0.00	-0.087	-0.108	-0.099	-0.03154	-0.06651	0.28259
794	4	0.597	3.13	-0.00	-0.093	-0.105	-0.108	-0.03363	-0.07171	0.28337
794	5	0.599	6.19	-0.00	-0.094	-0.105	-0.127	-0.03401	-0.07451	0.28541
794	6	0.599	9.28	-0.00	-0.114	-0.117	-0.154	-0.04108	-0.08701	0.29301
794	7	0.599	12.43	-0.00	-0.145	-0.150	-0.179	-0.05255	-0.10543	0.30714
794	8	0.598	-0.05	-0.00	-0.089	-0.110	-0.103	-0.03204	-0.06855	0.28220

TABLE VA
BALANCE CAVITY AND BASE PRESSURE COEFFICIENTS, AND INCREMENTAL
AXIAL FORCE COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	M _c	α	Φ	C _{p,c}	C _{p,b1}	C _{p,b2}	ΔC _{A,c}	ΔC _{A,b}	C _{A,u}
795	1	0.598	-3.07	-0.00	-0.088	-0.101	-0.098	-0.03168	-0.06385	0.27800
795	2	0.599	-0.01	-0.00	-0.086	-0.107	-0.099	-0.03096	-0.06644	0.28091
795	3	0.59A	1.02	-0.00	-0.086	-0.110	-0.099	-0.0310A	-0.06701	0.28259
795	4	0.59A	3.08	-0.00	-0.090	-0.109	-0.105	-0.03253	-0.06876	0.28657
795	5	0.59A	6.21	-0.00	-0.09E	-0.114	-0.128	-0.03534	-0.07761	0.28847
795	6	0.597	9.30	-0.00	-0.115	-0.121	-0.155	-0.04162	-0.08884	0.29690
795	7	0.597	12.46	-0.00	-0.146	-0.152	-0.186	-0.05262	-0.10848	0.31073
795	8	0.59A	-0.02	-0.00	-0.083	-0.104	-0.096	-0.02990	-0.06438	0.28175
796	1	0.599	-3.09	-0.00	-0.086	-0.097	-0.099	-0.03117	-0.06309	0.27739
796	2	0.599	-0.00	-0.00	-0.087	-0.108	-0.099	-0.03145	-0.06637	0.28225
796	3	0.599	0.99	-0.00	-0.08A	-0.110	-0.106	-0.03180	-0.06717	0.28634
796	4	0.599	3.10	-0.00	-0.090	-0.106	-0.106	-0.03256	-0.06831	0.29195
796	5	0.600	6.19	-0.00	-0.091	-0.104	-0.122	-0.03310	-0.07261	0.29537
796	6	0.600	9.28	-0.00	-0.10A	-0.113	-0.148	-0.03913	-0.08371	0.30309
796	7	0.600	12.43	-0.00	-0.137	-0.144	-0.172	-0.04932	-0.10142	0.31426
796	8	0.59A	-0.03	-0.00	-0.081	-0.104	-0.095	-0.02942	-0.06387	0.28129
797	1	0.59A	-3.07	-0.00	-0.085	-0.099	-0.103	-0.03091	-0.06487	0.28052
797	2	0.59A	-0.01	-0.00	-0.084	-0.109	-0.104	-0.03050	-0.06852	0.28949
797	3	0.59A	1.05	-0.00	-0.08A	-0.114	-0.100	-0.03171	-0.06881	0.29359
797	4	0.59A	3.12	-0.00	-0.092	-0.104	-0.107	-0.03312	-0.06803	0.30299
797	5	0.59A	6.21	-0.00	-0.096	-0.108	-0.124	-0.03488	-0.07437	0.30564
797	6	0.59A	9.31	-0.00	-0.112	-0.117	-0.152	-0.04041	-0.08650	0.31339
797	7	0.59A	12.43	-0.00	-0.138	-0.142	-0.178	-0.04982	-0.10296	0.32190
797	8	0.59A	-0.00	-0.00	-0.083	-0.111	-0.104	-0.02999	-0.06910	0.28924
798	1	0.598	-3.03	-0.00	-0.083	-0.097	-0.103	-0.02990	-0.06433	0.28652
798	2	0.599	-0.01	-0.00	-0.081	-0.110	-0.102	-0.02920	-0.06813	0.29932
798	3	0.59A	1.06	-0.00	-0.089	-0.117	-0.099	-0.03232	-0.06946	0.30112
798	4	0.59A	3.13	-0.00	-0.092	-0.108	-0.105	-0.03335	-0.06846	0.31365
798	5	0.59A	6.21	-0.00	-0.097	-0.106	-0.127	-0.03506	-0.07501	0.31832
798	6	0.598	9.28	-0.00	-0.109	-0.113	-0.152	-0.03942	-0.08519	0.32167
798	7	0.598	12.48	-0.00	-0.139	-0.148	-0.181	-0.05007	-0.10536	0.32741
798	8	0.59A	-0.02	-0.00	-0.083	-0.114	-0.105	-0.03013	-0.07047	0.30012
799	1	0.599	-3.03	-0.00	-0.085	-0.102	-0.114	-0.03093	-0.06941	0.30973
799	2	0.59A	-0.03	-0.00	-0.085	-0.127	-0.118	-0.03093	-0.07872	0.32941
799	3	0.599	1.08	-0.00	-0.083	-0.124	-0.102	-0.03016	-0.07267	0.32841
799	4	0.599	3.14	-0.00	-0.090	-0.112	-0.111	-0.03242	-0.07115	0.33851
799	5	0.599	6.22	-0.00	-0.107	-0.114	-0.149	-0.03385	-0.08428	0.33688

TABLE VA
BALANCE CAVITY AND BASE PRESSURE COEFFICIENTS, AND INCREMENTAL
AXIAL FORCE COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	M _c	α	φ	C _p c	C _p b ₁	C _p b ₂	ΔC _A c	ΔC _A b	C _A u
799	7	0.599	12.45	-0.00	-0.134	-0.140	-0.174	-0.04851	-0.10061	0.34265
799	8	0.599	0.05	-0.00	-0.082	-0.121	-0.116	-0.02955	-0.07617	0.32710
801	3	1.251	3.17	-0.00	-0.212	-0.213	-0.227	-0.07639	-0.14089	0.51949
801	4	1.250	-0.00	-0.00	-0.205	-0.218	-0.222	-0.07405	-0.14111	0.50934
801	5	1.251	1.00	-0.00	-0.205	-0.219	-0.223	-0.07415	-0.14181	0.50852
801	6	1.251	3.08	-0.00	-0.207	-0.217	-0.227	-0.07468	-0.14248	0.50858
801	7	1.251	6.22	-0.00	-0.226	-0.220	-0.226	-0.08169	-0.14926	0.54627
801	8	1.250	9.35	-0.00	-0.250	-0.238	-0.271	-0.09025	-0.16308	0.54627
801	9	1.252	12.60	-0.00	-0.273	-0.267	-0.292	-0.09857	-0.17913	0.56788
801	10	1.250	-0.07	-0.00	-0.208	-0.219	-0.226	-0.07506	-0.14268	0.50583
802	3	1.250	3.14	-0.00	-0.212	-0.215	-0.228	-0.07637	-0.14213	0.50660
802	4	1.250	-0.05	-0.00	-0.208	-0.220	-0.227	-0.07511	-0.14338	0.50185
802	5	1.251	1.01	-0.00	-0.211	-0.220	-0.226	-0.07492	-0.14306	0.50675
802	6	1.250	3.15	-0.00	-0.211	-0.221	-0.234	-0.07631	-0.14603	0.50939
802	7	1.250	6.26	-0.00	-0.228	-0.222	-0.249	-0.08241	-0.15096	0.52736
802	8	1.250	9.41	-0.00	-0.254	-0.239	-0.272	-0.09149	-0.16376	0.55287
802	9	1.249	12.63	-0.00	-0.278	-0.270	-0.297	-0.10011	-0.18209	0.57851
802	10	1.250	-0.02	-0.00	-0.208	-0.218	-0.226	-0.07505	-0.14219	0.50108
803	1	1.250	3.11	-0.00	-0.210	-0.214	-0.228	-0.07571	-0.14167	0.50481
803	2	1.250	-0.04	-0.00	-0.208	-0.220	-0.226	-0.07511	-0.14320	0.50301
803	3	1.249	1.02	-0.00	-0.210	-0.223	-0.229	-0.07574	-0.14487	0.50676
803	4	1.250	3.15	-0.00	-0.214	-0.223	-0.236	-0.07714	-0.14703	0.51340
803	5	1.249	6.29	-0.00	-0.231	-0.223	-0.251	-0.08348	-0.15203	0.53061
803	6	1.251	9.42	-0.00	-0.254	-0.240	-0.272	-0.09153	-0.16419	0.55595
803	7	1.251	12.62	-0.00	-0.278	-0.271	-0.297	-0.10037	-0.18190	0.58317
803	8	1.250	-0.04	-0.00	-0.211	-0.221	-0.228	-0.07604	-0.14395	0.50213
804	1	1.250	3.11	-0.00	-0.212	-0.216	-0.231	-0.07635	-0.14342	0.50443
804	2	1.250	-0.03	-0.00	-0.213	-0.226	-0.230	-0.07692	-0.14563	0.50866
804	3	1.250	1.05	-0.00	-0.213	-0.226	-0.232	-0.07688	-0.14686	0.51127
804	4	1.251	3.14	-0.00	-0.218	-0.225	-0.239	-0.07853	-0.14863	0.52010
804	5	1.250	6.31	-0.00	-0.231	-0.223	-0.253	-0.08338	-0.15247	0.53759
804	6	1.251	9.44	-0.00	-0.253	-0.238	-0.272	-0.09112	-0.16363	0.56428
804	7	1.250	12.64	-0.00	-0.278	-0.270	-0.298	-0.10042	-0.18207	0.59189
804	8	1.250	-0.00	-0.00	-0.211	-0.222	-0.229	-0.07629	-0.14459	0.50680
805	1	1.249	3.08	-0.00	-0.215	-0.221	-0.236	-0.07774	-0.14663	0.50988
805	2	1.249	-0.00	-0.00	-0.220	-0.232	-0.236	-0.07930	-0.14999	0.52155
805	3	1.249	1.08	-0.00	-0.217	-0.230	-0.235	-0.07813	-0.14931	0.52289

TABLE VA
 BALANCE CAVITY AND BASE PRESSURE COEFFICIENTS, AND INCREMENTAL
 AXIAL FORCE COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	M _c	α	Φ	C _p c	C _p b1	C _p b2	ΔC _A c	ΔC _A b	C _A u
805	4	1.250	3.20	-0.00	-0.224	-0.231	-0.246	-0.08083	-0.15295	0.53429
805	5	1.250	6.34	-0.00	-0.231	-0.224	-0.254	-0.08336	-0.15517	0.55022
805	6	1.250	9.47	-0.00	-0.229	-0.239	-0.272	-0.09154	-0.16384	0.57607
805	7	1.250	12.66	-0.00	-0.229	-0.239	-0.298	-0.10064	-0.18178	0.60508
805	8	1.250	0.03	-0.00	-0.220	-0.230	-0.235	-0.07947	-0.14911	0.52097
806	1	1.251	-3.05	-0.00	-0.219	-0.223	-0.238	-0.07885	-0.14795	0.52000
806	2	1.251	0.04	-0.00	-0.224	-0.234	-0.239	-0.08075	-0.15180	0.53670
806	3	1.250	1.09	-0.00	-0.223	-0.234	-0.239	-0.08028	-0.15180	0.53760
806	4	1.251	3.24	-0.00	-0.227	-0.233	-0.249	-0.08175	-0.15458	0.54786
806	5	1.252	6.36	-0.00	-0.230	-0.232	-0.253	-0.08313	-0.15250	0.56392
806	6	1.250	9.47	-0.00	-0.235	-0.238	-0.273	-0.09190	-0.16387	0.58878
806	7	1.252	12.65	-0.00	-0.280	-0.267	-0.297	-0.10080	-0.18082	0.61948
806	8	1.251	0.05	-0.00	-0.226	-0.235	-0.240	-0.08145	-0.15250	0.53555
807	1	1.250	-2.98	-0.00	-0.226	-0.230	-0.242	-0.08167	-0.15128	0.54820
807	2	1.249	0.09	-0.00	-0.231	-0.240	-0.244	-0.08335	-0.15499	0.56956
807	3	1.250	1.13	-0.00	-0.231	-0.240	-0.245	-0.08336	-0.15578	0.57666
807	4	1.251	3.25	-0.00	-0.228	-0.236	-0.251	-0.08388	-0.15643	0.59443
807	5	1.251	6.35	-0.00	-0.233	-0.225	-0.256	-0.08395	-0.15423	0.59443
807	6	1.251	9.48	-0.00	-0.232	-0.235	-0.271	-0.09075	-0.16243	0.61992
807	10	1.249	12.64	-0.00	-0.272	-0.263	-0.293	-0.09823	-0.17823	0.65407
807	11	1.252	0.07	-0.00	-0.226	-0.238	-0.241	-0.08149	-0.15356	0.57061
808	1	0.898	-3.02	-0.00	-0.115	-0.118	-0.145	-0.04150	-0.08457	0.32321
808	2	0.901	0.03	-0.00	-0.111	-0.140	-0.144	-0.04023	-0.09121	0.34617
808	3	0.901	1.07	-0.00	-0.118	-0.142	-0.135	-0.04270	-0.08896	0.34908
808	4	0.899	3.15	-0.00	-0.127	-0.140	-0.151	-0.04585	-0.09366	0.35206
808	5	0.901	6.24	-0.00	-0.130	-0.127	-0.157	-0.04690	-0.09138	0.35525
808	6	0.900	9.32	-0.00	-0.140	-0.135	-0.181	-0.05074	-0.10148	0.35723
808	7	0.899	12.42	-0.00	-0.183	-0.171	-0.226	-0.06610	-0.12739	0.38923
808	8	0.899	0.02	-0.00	-0.115	-0.141	-0.148	-0.04164	-0.09303	0.34601
809	1	0.899	-3.04	-0.00	-0.114	-0.117	-0.138	-0.04107	-0.08217	0.29861
809	2	0.899	0.06	-0.00	-0.117	-0.139	-0.136	-0.04088	-0.08832	0.31706
809	3	0.900	1.13	-0.00	-0.113	-0.137	-0.131	-0.04229	-0.08597	0.31996
809	4	0.899	3.21	-0.00	-0.126	-0.135	-0.148	-0.04544	-0.09086	0.33184
809	5	0.899	6.30	-0.00	-0.130	-0.126	-0.161	-0.04704	-0.09282	0.33404
809	6	0.899	9.47	-0.00	-0.147	-0.139	-0.188	-0.05318	-0.10486	0.35469
809	7	0.899	12.65	-0.00	-0.184	-0.172	-0.226	-0.06642	-0.12777	0.37039
809	8	0.899	0.03	-0.00	-0.112	-0.137	-0.136	-0.04051	-0.08781	0.31797

TABLE VA
 BALANCE CAVITY AND BASE PRESSURE COEFFICIENTS, AND INCREMENTAL
 AXIAL FORCE COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	M _c	α	Φ	C _{Pc}	C _{Pb1}	C _{Pb2}	ΔC _{Ac}	ΔC _{Ab}	C _{Au}
A10	1	0.898	-3.05	-0.00	-0.113	-0.115	-0.134	-0.04094	-0.07998	0.28962
A10	2	0.899	-0.01	-0.00	-0.113	-0.134	-0.131	-0.04072	-0.08500	0.30507
A10	3	0.899	1.04	-0.00	-0.121	-0.140	-0.135	-0.04359	-0.08822	0.30725
A10	4	0.89A	3.13	-0.00	-0.127	-0.132	-0.147	-0.04595	-0.08935	0.32074
A10	5	0.901	6.24	-0.00	-0.126	-0.125	-0.155	-0.04536	-0.08981	0.32381
A10	6	0.89A	9.30	-0.00	-0.146	-0.138	-0.188	-0.05263	-0.10455	0.32421
A10	7	0.900	12.46	-0.00	-0.181	-0.172	-0.224	-0.06533	-0.12685	0.336199
A10	8	0.899	-0.01	-0.00	-0.11A	-0.139	-0.138	-0.04263	-0.08906	0.30341
A11	1	0.89A	-3.08	-0.00	-0.114	-0.115	-0.135	-0.04110	-0.08019	0.28721
A11	2	0.899	-0.02	-0.00	-0.111	-0.127	-0.130	-0.04017	-0.08267	0.29575
A11	3	0.900	1.01	-0.00	-0.116	-0.133	-0.133	-0.04198	-0.08523	0.30023
A11	4	0.896	3.11	-0.00	-0.122	-0.132	-0.143	-0.04425	-0.08837	0.30707
A11	5	0.899	6.20	-0.00	-0.126	-0.127	-0.157	-0.04564	-0.09113	0.31149
A11	6	0.897	9.28	-0.00	-0.148	-0.140	-0.190	-0.05361	-0.10590	0.31683
A11	7	0.897	12.45	-0.00	-0.190	-0.176	-0.230	-0.06848	-0.13068	0.35146
A11	8	0.899	-0.02	-0.00	-0.115	-0.129	-0.132	-0.04140	-0.08362	0.29493
A12	1	0.89A	-3.09	-0.00	-0.117	-0.114	-0.135	-0.04227	-0.07999	0.29082
A12	2	0.89A	-0.03	-0.00	-0.114	-0.138	-0.131	-0.04106	-0.08336	0.29297
A12	3	0.89A	1.03	-0.00	-0.115	-0.131	-0.139	-0.04155	-0.08390	0.29346
A12	4	0.897	3.09	-0.00	-0.119	-0.128	-0.139	-0.04307	-0.08587	0.29883
A12	5	0.895	6.17	-0.00	-0.130	-0.132	-0.162	-0.04689	-0.09459	0.30355
A12	6	0.89A	9.27	-0.00	-0.146	-0.136	-0.188	-0.05277	-0.10455	0.31037
A12	7	0.900	12.47	-0.00	-0.190	-0.181	-0.230	-0.06869	-0.13163	0.34583
A12	8	0.901	-0.04	-0.00	-0.111	-0.126	-0.130	-0.04022	-0.08212	0.29188
A13	1	0.903	-3.07	-0.00	-0.112	-0.112	-0.128	-0.04047	-0.07719	0.29267
A13	2	0.899	-0.05	-0.00	-0.116	-0.130	-0.134	-0.04185	-0.08466	0.29107
A13	3	0.901	1.02	-0.00	-0.115	-0.132	-0.131	-0.04158	-0.08420	0.29348
A13	4	0.902	3.12	-0.00	-0.117	-0.126	-0.135	-0.04225	-0.08377	0.29630
A13	5	0.899	6.19	-0.00	-0.125	-0.123	-0.158	-0.04529	-0.09160	0.30146
A13	6	0.899	9.31	-0.00	-0.149	-0.143	-0.191	-0.05382	-0.10721	0.30777
A13	7	0.899	12.43	-0.00	-0.191	-0.181	-0.231	-0.06904	-0.13208	0.34260
A13	8	0.901	-0.03	-0.00	-0.113	-0.126	-0.131	-0.04075	-0.08267	0.29186
A14	1	0.899	-3.13	-0.00	-0.121	-0.119	-0.136	-0.04376	-0.08166	0.30307
A14	2	0.900	-0.06	-0.00	-0.112	-0.126	-0.135	-0.04060	-0.08378	0.29428
A14	3	0.900	1.09	-0.00	-0.115	-0.131	-0.132	-0.04144	-0.08516	0.29113
A14	4	0.900	3.10	-0.00	-0.116	-0.130	-0.135	-0.04205	-0.08516	0.29428
A14	5	0.902	6.17	-0.00	-0.123	-0.128	-0.154	-0.04431	-0.09054	0.29466
A14	6	0.899	9.28	-0.00	-0.152	-0.144	-0.192	-0.05475	-0.10796	0.29090
A14	7	0.901	12.45	-0.00	-0.183	-0.174	-0.223	-0.06620	-0.12733	0.33568

TABLE VA
 BALANCE CAVITY AND BASE PRESSURE COEFFICIENTS, AND INCREMENTAL
 AXIAL FORCE COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	M _C	α	Φ	C _{Pc}	C _{Pbl}	C _{Pb2}	ΔC _{Ac}	ΔC _{Ab}	C _{Au}
A14	8	0.902	-0.07	-0.00	-0.111	-0.125	-0.132	-0.04018	-0.08247	0.29406
A15	7	0.600	-3.10	-0.00	-0.091	-0.098	-0.098	-0.03305	-0.06282	0.28821
A15	8	0.600	-0.05	-0.00	-0.086	-0.108	-0.103	-0.03131	-0.06777	0.28318
A15	9	0.600	0.98	-0.00	-0.088	-0.114	-0.098	-0.03175	-0.06829	0.28351
A15	10	0.600	3.07	-0.00	-0.089	-0.113	-0.103	-0.03233	-0.06963	0.28090
A15	11	0.600	6.18	-0.00	-0.095	-0.108	-0.121	-0.03450	-0.07374	0.28068
A15	12	0.600	9.27	-0.00	-0.116	-0.120	-0.155	-0.04205	-0.08836	0.28681
A15	13	0.600	12.44	-0.00	-0.144	-0.151	-0.181	-0.05213	-0.10637	0.29850
A15	14	0.600	-0.04	-0.00	-0.087	-0.111	-0.102	-0.03136	-0.06869	0.28227
A16	1	0.600	-3.09	-0.00	-0.090	-0.099	-0.102	-0.03269	-0.06446	0.27946
A16	2	0.600	-0.02	-0.00	-0.085	-0.108	-0.099	-0.03092	-0.06663	0.28133
A16	3	0.600	0.99	-0.00	-0.088	-0.112	-0.099	-0.03177	-0.06800	0.28266
A16	4	0.601	3.08	-0.00	-0.089	-0.110	-0.103	-0.03226	-0.06854	0.28418
A16	5	0.600	6.19	-0.00	-0.095	-0.106	-0.124	-0.03455	-0.07383	0.28407
A16	6	0.600	9.27	-0.00	-0.116	-0.119	-0.154	-0.04176	-0.08785	0.29379
A16	7	0.600	12.45	-0.00	-0.143	-0.150	-0.183	-0.05158	-0.10668	0.30691
A16	8	0.600	-0.01	-0.00	-0.085	-0.107	-0.099	-0.03076	-0.06619	0.27998
A17	1	0.601	-3.08	-0.00	-0.085	-0.097	-0.097	-0.03077	-0.06216	0.27829
A17	2	0.599	-0.02	-0.00	-0.089	-0.110	-0.100	-0.03228	-0.06771	0.28230
A17	3	0.599	1.00	-0.00	-0.089	-0.112	-0.102	-0.03228	-0.06893	0.28391
A17	4	0.600	3.10	-0.00	-0.089	-0.108	-0.102	-0.03223	-0.06737	0.28544
A17	5	0.600	6.22	-0.00	-0.100	-0.112	-0.128	-0.03611	-0.07698	0.28734
A17	6	0.599	9.28	-0.00	-0.117	-0.120	-0.156	-0.04239	-0.08868	0.29702
A17	7	0.599	12.45	-0.00	-0.146	-0.152	-0.184	-0.05279	-0.10776	0.31027
A17	8	0.600	-0.04	-0.00	-0.085	-0.106	-0.098	-0.03063	-0.06564	0.28095
A18	1	0.599	-3.06	-0.00	-0.089	-0.107	-0.102	-0.03227	-0.06550	0.27837
A18	2	0.600	-0.04	-0.00	-0.086	-0.110	-0.102	-0.03098	-0.06705	0.28277
A18	3	0.600	1.03	-0.00	-0.090	-0.110	-0.101	-0.03251	-0.06789	0.28562
A18	4	0.600	3.13	-0.00	-0.091	-0.108	-0.104	-0.03303	-0.06802	0.29016
A18	5	0.600	6.21	-0.00	-0.096	-0.107	-0.126	-0.03481	-0.07480	0.29527
A18	6	0.600	9.28	-0.00	-0.115	-0.116	-0.153	-0.04145	-0.08636	0.30273
A18	7	0.599	12.46	-0.00	-0.144	-0.149	-0.179	-0.05186	-0.10543	0.31584
A18	8	0.600	-0.02	-0.00	-0.086	-0.107	-0.095	-0.03104	-0.06496	0.28166
A19	1	0.600	-3.05	-0.00	-0.087	-0.096	-0.101	-0.03147	-0.06324	0.28010
A19	2	0.600	1.00	-0.00	-0.085	-0.110	-0.102	-0.03064	-0.06810	0.28974
A19	3	0.600	3.03	-0.00	-0.085	-0.110	-0.100	-0.03225	-0.06909	0.29301
A19	4	0.599	6.12	-0.00	-0.093	-0.108	-0.110	-0.03355	-0.07005	0.30184

TABLE VA
 BALANCE CAVITY AND BASE PRESSURE COEFFICIENTS, AND INCREMENTAL
 AXIAL FORCE COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	M _c	α	φ	C _p c	C _p b ₁	C _p b ₂	ΔC _A c	ΔC _A b	C _A u
819	5	0.599	6.21	-0.00	-0.099	-0.107	-0.124	-0.03584	-0.07444	0.30666
A19	6	0.598	9.34	-0.00	-0.116	-0.119	-0.156	-0.04202	-0.08853	0.314023
A19	7	0.599	12.44	-0.00	-0.142	-0.146	-0.182	-0.05145	-0.10520	0.32423
A19	8	0.599	-0.00	-0.00	-0.085	-0.112	-0.106	-0.03078	-0.07012	0.28974
A20	1	0.600	-3.05	-0.00	-0.087	-0.098	-0.104	-0.03149	-0.06507	0.28509
A20	2	0.599	-0.00	-0.00	-0.098	-0.123	-0.113	-0.03199	-0.07569	0.30112
A20	3	0.600	1.04	-0.00	-0.090	-0.119	-0.100	-0.03271	-0.07041	0.30133
A20	4	0.600	3.14	-0.00	-0.093	-0.108	-0.113	-0.03380	-0.07116	0.31252
A20	5	0.600	6.23	-0.00	-0.099	-0.108	-0.126	-0.03582	-0.07520	0.31752
A20	6	0.600	9.31	-0.00	-0.112	-0.116	-0.149	-0.04052	-0.08537	0.32210
A20	7	0.601	12.44	-0.00	-0.140	-0.145	-0.181	-0.05055	-0.10451	0.32905
A20	8	0.600	0.01	-0.00	-0.084	-0.117	-0.105	-0.03042	-0.07125	0.30063
A21	1	0.599	-3.00	-0.00	-0.086	-0.101	-0.109	-0.03108	-0.06754	0.30778
A21	2	0.600	0.01	-0.00	-0.086	-0.125	-0.118	-0.03108	-0.07806	0.32958
A21	3	0.599	1.06	-0.00	-0.086	-0.126	-0.102	-0.03115	-0.07312	0.32929
A21	4	0.599	3.18	-0.00	-0.097	-0.114	-0.118	-0.03523	-0.07467	0.33645
A21	5	0.599	6.23	-0.00	-0.101	-0.109	-0.127	-0.03678	-0.07574	0.33944
A21	6	0.599	9.31	-0.00	-0.112	-0.118	-0.155	-0.04049	-0.08788	0.33849
A21	7	0.600	12.44	-0.00	-0.135	-0.136	-0.175	-0.04878	-0.09989	0.34190
A21	8	0.601	0.02	-0.00	-0.086	-0.121	-0.116	-0.03125	-0.07639	0.32776

TABLE VA
BALANCE CAVITY AND BASE PRESSURE COEFFICIENTS, AND INCREMENTAL
AXIAL FORCE COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	M _c	α	φ	C _{pC}	C _{pbl}	C _{pbl2}	ΔC _{Ac}	ΔC _{Ab}	C _{Au}
A22	3	0.901	-3.09	-0.00	-0.121	-0.127	-0.141	-0.04366	-0.08611	0.30352
A22	4	0.901	-0.07	-0.00	-0.123	-0.135	-0.139	-0.04460	-0.08797	0.29886
A22	5	0.900	1.00	-0.00	-0.121	-0.134	-0.136	-0.04361	-0.08672	0.29512
A22	6	0.902	3.10	-0.00	-0.122	-0.134	-0.134	-0.04418	-0.08616	0.29536
A22	7	0.898	4.16	-0.00	-0.130	-0.145	-0.145	-0.04693	-0.09304	0.29159
A22	8	0.898	9.27	-0.00	-0.152	-0.170	-0.169	-0.05484	-0.10901	0.29562
A22	9	0.900	12.46	-0.00	-0.169	-0.179	-0.217	-0.06114	-0.12684	0.29192
A22	10	0.900	-0.06	-0.00	-0.126	-0.138	-0.143	-0.04546	-0.09016	0.29697
A23	1	0.901	-3.09	-0.00	-0.124	-0.128	-0.146	-0.04494	-0.08790	0.29448
A23	3	0.899	-0.03	-0.00	-0.126	-0.134	-0.140	-0.04412	-0.08807	0.29216
A23	4	0.902	3.13	-0.00	-0.122	-0.134	-0.136	-0.04403	-0.09125	0.29471
A23	5	0.899	6.19	-0.00	-0.135	-0.150	-0.152	-0.04878	-0.09713	0.29853
A23	6	0.899	9.31	-0.00	-0.148	-0.166	-0.164	-0.05344	-0.10620	0.30067
A23	7	0.901	12.47	-0.00	-0.146	-0.164	-0.160	-0.05260	-0.10418	0.30059
A23	8	0.901	-0.01	-0.00	-0.120	-0.133	-0.212	-0.05985	-0.12349	0.29868
A23	9	0.901	-0.01	-0.00	-0.120	-0.133	-0.139	-0.04323	-0.08761	0.29166
A24	1	0.898	-0.02	-0.00	-0.123	-0.137	-0.144	-0.04459	-0.09025	0.29292
A24	2	0.897	-3.05	-0.00	-0.124	-0.130	-0.148	-0.04470	-0.08934	0.29187
A24	3	0.897	-0.02	-0.00	-0.124	-0.138	-0.146	-0.04491	-0.09115	0.29099
A24	4	0.900	1.00	-0.00	-0.120	-0.134	-0.140	-0.04354	-0.08818	0.29580
A24	5	0.899	3.13	-0.00	-0.122	-0.134	-0.138	-0.04403	-0.08747	0.29910
A24	6	0.897	6.21	-0.00	-0.138	-0.154	-0.155	-0.04984	-0.09937	0.29960
A24	7	0.898	9.30	-0.00	-0.151	-0.172	-0.169	-0.05464	-0.10937	0.30119
A24	8	0.897	12.46	-0.00	-0.175	-0.184	-0.223	-0.06319	-0.13056	0.29882
A24	9	0.900	-0.01	-0.00	-0.124	-0.138	-0.145	-0.04476	-0.09067	0.29285
A25	3	0.899	-3.05	-0.00	-0.127	-0.130	-0.152	-0.04588	-0.09050	0.29204
A25	4	0.898	-0.02	-0.00	-0.124	-0.136	-0.147	-0.04485	-0.09095	0.29716
A25	5	0.898	1.05	-0.00	-0.124	-0.136	-0.145	-0.04470	-0.09033	0.30279
A25	6	0.898	3.12	-0.00	-0.126	-0.138	-0.140	-0.04564	-0.09221	0.30590
A25	7	0.898	6.21	-0.00	-0.136	-0.151	-0.151	-0.04923	-0.09673	0.30788
A25	8	0.898	9.30	-0.00	-0.149	-0.162	-0.164	-0.05369	-0.10473	0.30730
A25	9	0.900	12.47	-0.00	-0.168	-0.175	-0.214	-0.06082	-0.12477	0.30643
A25	10	0.899	0.00	-0.00	-0.127	-0.139	-0.150	-0.04590	-0.09277	0.29435
A26	1	0.898	-3.03	-0.00	-0.121	-0.126	-0.150	-0.04378	-0.08955	0.29090
A26	2	0.898	-0.04	-0.00	-0.127	-0.140	-0.150	-0.04580	-0.09329	0.30580
A26	3	0.898	1.02	-0.00	-0.135	-0.148	-0.158	-0.04865	-0.09824	0.31290
A26	4	0.900	3.12	-0.00	-0.129	-0.139	-0.142	-0.04661	-0.09034	0.31696

TABLE VA
 BALANCE CAVITY AND BASE PRESSURE COEFFICIENTS, AND INCREMENTAL
 AXIAL FORCE COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	M _c	α	Φ	C _p c	C _p b ₁	C _p b ₂	ΔC _A c	ΔC _A b	C _A u
A26	5	0.902	6.22	-0.00	-0.133	-0.147	-0.146	-0.04810	-0.09394	0.31841
A26	6	0.901	9.31	-0.00	-0.149	-0.161	-0.162	-0.05366	-0.10391	0.31600
A26	7	0.899	12.46	-0.00	-0.170	-0.175	-0.214	-0.06150	-0.12470	0.31609
A26	8	0.89A	0.00	-0.00	-0.129	-0.142	-0.153	-0.04675	-0.09458	0.30412
A27	1	0.898	-3.05	-0.00	-0.130	-0.134	-0.159	-0.04714	-0.09414	0.30054
A27	2	0.89A	0.04	-0.00	-0.135	-0.150	-0.158	-0.04861	-0.09862	0.31842
A27	3	0.900	1.06	-0.00	-0.138	-0.152	-0.160	-0.04974	-0.09995	0.32931
A27	4	0.899	3.16	-0.00	-0.134	-0.143	-0.147	-0.04855	-0.09269	0.32654
A27	5	0.899	6.23	-0.00	-0.140	-0.153	-0.152	-0.05044	-0.09805	0.32623
A27	6	0.899	9.31	-0.00	-0.151	-0.162	-0.163	-0.05444	-0.10451	0.32563
A27	7	0.901	12.48	-0.00	-0.167	-0.166	-0.209	-0.06015	-0.12058	0.32489
A27	8	0.902	0.02	-0.00	-0.131	-0.146	-0.154	-0.04730	-0.09658	0.32025
A28	1	0.900	-3.00	-0.00	-0.129	-0.132	-0.155	-0.04650	-0.09239	0.32481
A28	2	0.900	0.07	-0.00	-0.135	-0.151	-0.158	-0.04880	-0.09923	0.33835
A28	3	0.902	1.08	-0.00	-0.135	-0.149	-0.156	-0.04870	-0.09783	0.35596
A28	4	0.899	3.19	-0.00	-0.131	-0.139	-0.148	-0.04728	-0.09227	0.35358
A28	5	0.89A	6.23	-0.00	-0.138	-0.150	-0.149	-0.04972	-0.09581	0.34558
A28	6	0.89A	9.32	-0.00	-0.149	-0.161	-0.160	-0.05372	-0.10305	0.34669
A28	7	0.899	12.49	-0.00	-0.171	-0.168	-0.211	-0.06164	-0.12145	0.34254
A28	8	0.899	0.05	-0.00	-0.136	-0.148	-0.157	-0.0489A	-0.09781	0.33862
A29	3	0.898	-3.14	44.99	-0.123	-0.145	-0.142	-0.04449	-0.09236	0.30948
A29	4	0.899	0.00	44.99	-0.121	-0.135	-0.139	-0.04376	-0.08818	0.29712
A29	5	0.89A	1.00	44.99	-0.123	-0.134	-0.141	-0.04347	-0.08838	0.29238
A29	6	0.896	3.06	44.99	-0.120	-0.133	-0.140	-0.04353	-0.08774	0.29253
A29	7	0.896	6.16	44.99	-0.12A	-0.140	-0.152	-0.04637	-0.09379	0.29709
A29	8	0.897	9.31	44.99	-0.139	-0.149	-0.194	-0.05037	-0.12354	0.30962
A29	9	0.897	12.42	44.99	-0.155	-0.124	-0.213	-0.05599	-0.14004	0.31131
A29	10	0.895	0.03	44.99	-0.191	-0.271	-0.271	-0.06891	-0.17370	0.34607
A30	1	0.897	-3.09	44.99	-0.122	-0.134	-0.140	-0.04407	-0.08797	0.29950
A30	2	0.897	0.00	44.99	-0.121	-0.135	-0.139	-0.04376	-0.08818	0.29238
A30	3	0.897	-0.00	44.99	-0.123	-0.134	-0.141	-0.04347	-0.08838	0.29253
A30	4	0.89A	1.03	44.99	-0.120	-0.133	-0.140	-0.04353	-0.08774	0.29507
A30	5	0.895	3.09	44.99	-0.12A	-0.140	-0.152	-0.04637	-0.09379	0.29709
A30	6	0.896	6.21	44.99	-0.139	-0.149	-0.194	-0.05037	-0.12354	0.30962
A30	7	0.896	9.33	44.99	-0.155	-0.124	-0.213	-0.05599	-0.14004	0.31131
A30	8	0.896	12.44	44.99	-0.191	-0.271	-0.271	-0.06891	-0.17370	0.34607
A30	9	0.895	0.01	44.99	-0.122	-0.134	-0.140	-0.04407	-0.08797	0.29950
A31	1	0.897	-3.08	44.99	-0.124	-0.147	-0.145	-0.04482	-0.09362	0.29463

TABLE VA
BALANCE CAVITY AND BASE PRESSURE COEFFICIENTS, AND INCREMENTAL AXIAL FORCE COEFFICIENTS - AERODYNAMIC CASE

Run	Pt.	M _c	α	Φ	C _{Pc}	C _{Pb1}	C _{Pb2}	ΔC _{Ac}	ΔC _{Ab}	C _{Au}
A31	2	0.899	-0.00	44.99	-0.118	-0.133	-0.134	-0.04254	-0.08595	0.29431
A31	3	0.89A	1.04	44.99	-0.119	-0.133	-0.137	-0.04297	-0.08653	0.29618
A31	4	0.89A	3.10	44.99	-0.132	-0.142	-0.158	-0.04775	-0.09613	0.30299
A31	5	0.89A	6.22	44.99	-0.139	-0.149	-0.194	-0.05024	-0.12439	0.31376
A31	6	0.89A	9.35	44.99	-0.151	-0.220	-0.208	-0.05459	-0.13739	0.31707
A31	7	0.900	12.45	45.00	-0.185	-0.266	-0.265	-0.06664	-0.17046	0.34808
A31	8	0.897	-0.00	44.99	-0.121	-0.137	-0.140	-0.0438A	-0.08883	0.29367
A32	1	0.897	-3.06	44.99	-0.123	-0.146	-0.142	-0.04449	-0.09242	0.29327
A32	2	0.897	0.03	44.99	-0.12A	-0.152	-0.147	-0.04631	-0.09586	0.29252
A32	3	0.897	1.06	44.99	-0.125	-0.145	-0.143	-0.04519	-0.09242	0.30862
A32	4	0.896	3.11	44.99	-0.129	-0.141	-0.155	-0.04671	-0.09506	0.31354
A32	5	0.896	6.26	44.99	-0.135	-0.190	-0.188	-0.04864	-0.12106	0.32473
A32	6	0.896	9.37	44.99	-0.157	-0.225	-0.214	-0.05681	-0.14077	0.32878
A32	7	0.89A	12.47	45.00	-0.1A8	-0.268	-0.266	-0.06768	-0.17136	0.35878
A32	8	0.896	0.02	44.99	-0.12A	-0.150	-0.147	-0.04643	-0.09532	0.29992
A33	1	0.896	-3.05	44.99	-0.121	-0.146	-0.14A	-0.04391	-0.09447	0.29960
A33	2	0.899	0.06	44.99	-0.125	-0.152	-0.145	-0.04518	-0.09558	0.22250
A33	3	0.89A	1.10	44.99	-0.138	-0.162	-0.157	-0.04983	-0.10263	0.33474
A33	4	0.89A	3.16	44.99	-0.139	-0.160	-0.147	-0.04528	-0.09233	0.33745
A33	5	0.89A	6.26	44.99	-0.139	-0.193	-0.192	-0.05037	-0.12333	0.34781
A33	6	0.897	9.40	44.99	-0.153	-0.220	-0.208	-0.0553A	-0.13833	0.34686
A33	7	0.899	12.49	45.00	-0.191	-0.270	-0.271	-0.06879	-0.17339	0.37953
A33	8	0.899	0.05	44.99	-0.12A	-0.153	-0.149	-0.04632	-0.09698	0.22223
A34	1	0.895	-3.02	44.99	-0.125	-0.148	-0.152	-0.04529	-0.09650	0.31847
A34	2	0.895	0.02	44.99	-0.130	-0.158	-0.156	-0.04686	-0.10072	0.34630
A34	3	0.895	1.17	44.99	-0.141	-0.169	-0.166	-0.05093	-0.10753	0.35726
A34	4	0.897	3.28	44.99	-0.134	-0.167	-0.153	-0.04848	-0.09939	0.36370
A34	5	0.895	6.28	44.99	-0.140	-0.189	-0.192	-0.05062	-0.12216	0.36941
A34	6	0.897	9.41	44.99	-0.148	-0.217	-0.203	-0.05363	-0.13450	0.40316
A34	7	0.897	12.50	45.00	-0.189	-0.265	-0.263	-0.06805	-0.16928	0.43456
A34	8	0.896	0.0A	44.99	-0.133	-0.159	-0.159	-0.04819	-0.10219	0.34556
A35	1	0.895	-2.96	44.99	-0.128	-0.142	-0.157	-0.04406	-0.09606	0.37110
A35	2	0.895	0.12	44.99	-0.128	-0.154	-0.157	-0.04629	-0.09975	0.38774
A35	3	0.896	1.15	44.99	-0.138	-0.168	-0.164	-0.04995	-0.10652	0.40871
A35	4	0.897	3.21	44.99	-0.143	-0.171	-0.159	-0.05174	-0.10605	0.42577
A35	5	0.896	6.30	44.99	-0.145	-0.185	-0.182	-0.05446	-0.11757	0.41225
A35	6	0.897	9.43	44.99	-0.145	-0.211	-0.198	-0.0524A	-0.13097	0.44189
A35	7	0.897	12.51	45.00	-0.182	-0.257	-0.253	-0.06562	-0.16344	0.44995

TABLE VA
BALANCE CAVITY AND BASE PRESSURE COEFFICIENTS, AND INCREMENTAL AXIAL FORCE COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	M _c	α	Φ	C _p c	C _p b ₁	C _p b ₂	ΔC _A c	ΔC _A b	C _A u
A35	8	0.897	0.11	44.99	-0.130	-0.154	-0.159	-0.04694	-0.10037	0.38777
A36	5	1.250	-3.17	-0.00	-0.228	-0.235	-0.245	-0.08224	-0.15399	0.52426
A36	6	1.250	-0.05	-0.00	-0.220	-0.228	-0.237	-0.07921	-0.14903	0.50672
A36	7	1.250	0.99	-0.00	-0.218	-0.227	-0.236	-0.07877	-0.14849	0.50829
A36	8	1.250	3.08	-0.00	-0.223	-0.232	-0.242	-0.08059	-0.15183	0.51237
A36	9	1.248	6.27	-0.00	-0.245	-0.245	-0.261	-0.08849	-0.16243	0.52443
A36	10	1.248	9.38	-0.00	-0.261	-0.267	-0.282	-0.09420	-0.17604	0.54307
A36	11	1.249	12.60	-0.00	-0.284	-0.280	-0.300	-0.10255	-0.18600	0.55648
A36	12	1.250	-0.05	-0.00	-0.219	-0.228	-0.238	-0.07918	-0.14943	0.50452
A37	1	1.251	-3.18	-0.00	-0.225	-0.233	-0.243	-0.08115	-0.15261	0.51384
A37	2	1.250	-0.01	-0.00	-0.220	-0.231	-0.239	-0.07943	-0.15058	0.50139
A37	3	1.252	0.99	-0.00	-0.227	-0.228	-0.237	-0.07929	-0.14903	0.50816
A37	4	1.248	3.14	-0.00	-0.223	-0.233	-0.246	-0.08207	-0.15470	0.51602
A37	5	1.248	6.28	-0.00	-0.243	-0.243	-0.260	-0.08769	-0.16115	0.52659
A37	6	1.249	9.41	-0.00	-0.263	-0.262	-0.280	-0.09481	-0.17369	0.54739
A37	7	1.251	12.66	-0.00	-0.280	-0.277	-0.297	-0.10090	-0.18399	0.56557
A37	8	1.249	-0.02	-0.00	-0.219	-0.228	-0.238	-0.07914	-0.14942	0.50994
A38	1	1.250	3.3	-0.00	-0.225	-0.233	-0.244	-0.08101	-0.15289	0.50990
A38	2	1.249	-0.00	-0.00	-0.218	-0.229	-0.237	-0.07866	-0.14966	0.50204
A38	3	1.250	0.33	-0.00	-0.221	-0.231	-0.239	-0.07976	-0.15064	0.50922
A38	4	1.250	3.19	-0.00	-0.240	-0.240	-0.246	-0.08190	-0.15472	0.51800
A38	5	1.250	6.40	-0.00	-0.261	-0.260	-0.277	-0.08640	-0.17152	0.52888
A38	6	1.251	9.40	-0.00	-0.280	-0.276	-0.296	-0.09406	-0.18337	0.54980
A38	7	1.251	12.65	-0.00	-0.281	-0.278	-0.296	-0.10094	-0.18337	0.56871
A38	8	1.250	-0.03	-0.00	-0.217	-0.228	-0.237	-0.07822	-0.14911	0.50061
A39	1	1.250	3.09	-0.00	-0.224	-0.233	-0.246	-0.08074	-0.15362	0.50638
A39	2	1.250	-0.03	-0.00	-0.219	-0.232	-0.240	-0.07914	-0.15153	0.50546
A39	3	1.250	3.18	-0.00	-0.224	-0.234	-0.242	-0.08071	-0.15257	0.51539
A39	4	1.251	6.42	-0.00	-0.230	-0.237	-0.247	-0.08284	-0.15502	0.52594
A39	5	1.250	9.42	-0.00	-0.239	-0.241	-0.260	-0.08638	-0.16061	0.53570
A39	6	1.250	12.64	-0.00	-0.262	-0.256	-0.276	-0.09439	-0.17077	0.55569
A39	7	1.247	-0.00	-0.00	-0.284	-0.282	-0.304	-0.10231	-0.18810	0.57749
A39	8	1.254	-0.02	-0.00	-0.216	-0.229	-0.237	-0.07798	-0.14943	0.50628
A40	1	1.250	-3.08	-0.00	-0.224	-0.235	-0.251	-0.08076	-0.15591	0.51022
A40	2	1.241	0.01	-0.00	-0.228	-0.244	-0.253	-0.08231	-0.15945	0.51430
A40	3	1.249	3.01	-0.00	-0.224	-0.239	-0.247	-0.08097	-0.15604	0.51581
A40	4	1.250	6.08	-0.00	-0.228	-0.240	-0.247	-0.08236	-0.15636	0.52741

TABLE VA
BALANCE CAVITY AND BASE PRESSURE COEFFICIENTS, AND INCREMENTAL
AXIAL FORCE COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	M _c	α	φ	C _{p_c}	C _{p_{b1}}	C _{p_{b2}}	ΔC _{A_c}	ΔC _{A_b}	C _{A_u}
340	5	1.250	3.19	-0.00	-0.234	-0.240	-0.251	-0.08429	-0.15759	0.53897
340	6	1.251	6.33	-0.00	-0.237	-0.241	-0.258	-0.08548	-0.16010	0.54919
340	7	1.250	9.45	-0.00	-0.256	-0.251	-0.274	-0.09229	-0.16856	0.56629
340	8	1.250	12.65	-0.00	-0.280	-0.273	-0.295	-0.10087	-0.18207	0.58880
340	9	1.250	0.01	-0.00	-0.225	-0.239	-0.247	-0.08109	-0.15595	0.51459
341	1	1.251	-3.05	-0.00	-0.228	-0.238	-0.256	-0.08232	-0.15849	0.51900
341	2	1.250	0.05	-0.00	-0.227	-0.242	-0.250	-0.08185	-0.15767	0.52934
341	6	1.249	1.10	-0.00	-0.229	-0.242	-0.249	-0.08268	-0.15743	0.54034
341	7	1.251	3.21	-0.00	-0.234	-0.239	-0.250	-0.08426	-0.15677	0.55400
341	8	1.252	6.35	-0.00	-0.234	-0.240	-0.254	-0.08459	-0.15838	0.56510
341	9	1.251	9.46	-0.00	-0.254	-0.248	-0.269	-0.09166	-0.16578	0.58114
341	10	1.251	12.66	-0.00	-0.275	-0.270	-0.290	-0.09916	-0.17952	0.60475
341	11	1.250	0.04	-0.00	-0.223	-0.238	-0.246	-0.08038	-0.15518	0.52766
342	1	1.250	-3.01	-0.00	-0.226	-0.237	-0.253	-0.08140	-0.15729	0.54529
342	2	1.249	0.09	-0.00	-0.229	-0.242	-0.249	-0.08103	-0.15730	0.55874
342	3	1.249	1.12	-0.00	-0.229	-0.244	-0.252	-0.08270	-0.15899	0.57116
342	4	1.248	3.24	-0.00	-0.234	-0.245	-0.253	-0.08451	-0.15965	0.58886
342	5	1.249	6.36	-0.00	-0.236	-0.249	-0.255	-0.08508	-0.15948	0.59434
342	6	1.250	9.48	-0.00	-0.253	-0.243	-0.269	-0.09143	-0.16606	0.61377
342	7	1.250	12.68	-0.00	-0.274	-0.266	-0.290	-0.09890	-0.17830	0.63347
342	8	1.250	0.04	-0.00	-0.223	-0.240	-0.248	-0.08036	-0.15631	0.55746
343	3	1.250	-3.21	44.99	-0.237	-0.246	-0.252	-0.08565	-0.15941	0.53893
343	4	1.249	0.09	44.99	-0.220	-0.231	-0.239	-0.07944	-0.15094	0.50870
343	5	1.250	0.98	44.99	-0.215	-0.226	-0.236	-0.07756	-0.14793	0.50259
343	6	1.249	3.07	44.99	-0.220	-0.232	-0.242	-0.07943	-0.15180	0.50517
343	7	1.250	6.23	44.99	-0.242	-0.269	-0.269	-0.08718	-0.17248	0.52562
343	8	1.250	9.40	44.99	-0.260	-0.291	-0.282	-0.09392	-0.18384	0.54231
343	9	1.249	12.57	44.99	-0.280	-0.305	-0.301	-0.10098	-0.19420	0.55850
343	10	1.251	0.05	44.99	-0.216	-0.228	-0.237	-0.07809	-0.14926	0.50788
344	1	1.250	-3.16	44.99	-0.226	-0.235	-0.245	-0.08150	-0.15395	0.51467
344	2	1.251	0.03	44.99	-0.214	-0.224	-0.234	-0.07709	-0.14679	0.49990
344	3	1.251	1.01	44.99	-0.217	-0.226	-0.235	-0.07816	-0.14755	0.50243
344	4	1.250	3.11	44.99	-0.226	-0.235	-0.245	-0.08164	-0.15404	0.51423
344	5	1.249	6.26	44.99	-0.238	-0.272	-0.270	-0.08600	-0.17372	0.53212
344	6	1.249	9.44	44.99	-0.259	-0.292	-0.282	-0.09332	-0.18418	0.55309
344	7	1.249	12.60	44.99	-0.280	-0.307	-0.301	-0.10102	-0.19463	0.57259
344	8	1.249	0.03	44.99	-0.216	-0.225	-0.235	-0.07778	-0.14759	0.50475

TABLE VA
BALANCE CAVITY AND BASE PRESSURE COEFFICIENTS, AND INCREMENTAL
AXIAL FORCE COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	M _c	α	Φ	C _{p,c}	C _{p,b1}	C _{p,b2}	ΔC _{A,c}	ΔC _{A,b}	C _{A,u}
345	1	1.250	-3.16	44.99	-0.223	-0.232	-0.243	-0.080	-0.152	0.508
345	2	1.251	-0.01	44.99	-0.214	-0.224	-0.233	-0.077	-0.146	0.500
345	3	1.251	1.03	44.99	-0.218	-0.229	-0.237	-0.076	-0.149	0.507
345	4	1.251	3.13	44.99	-0.231	-0.239	-0.248	-0.083	-0.156	0.521
345	5	1.249	6.27	44.99	-0.239	-0.273	-0.273	-0.086	-0.174	0.536
345	6	1.251	9.44	44.99	-0.257	-0.291	-0.280	-0.092	-0.183	0.557
345	7	1.251	12.61	44.99	-0.274	-0.304	-0.298	-0.100	-0.192	0.578
345	8	1.250	-0.03	44.99	-0.214	-0.223	-0.233	-0.077	-0.146	0.509
346	16	1.251	-3.14	44.99	-0.217	-0.225	-0.235	-0.078	-0.147	0.504
346	17	1.249	0.00	44.99	-0.219	-0.230	-0.235	-0.079	-0.149	0.507
346	18	1.249	1.05	44.99	-0.228	-0.236	-0.244	-0.082	-0.153	0.519
346	19	1.250	3.16	44.99	-0.237	-0.244	-0.253	-0.085	-0.159	0.537
346	20	1.250	6.29	44.99	-0.235	-0.262	-0.270	-0.084	-0.170	0.547
346	21	1.250	9.46	44.99	-0.254	-0.289	-0.278	-0.091	-0.181	0.568
346	22	1.251	12.63	44.99	-0.275	-0.302	-0.294	-0.099	-0.190	0.591
346	23	1.251	0.00	44.99	-0.216	-0.227	-0.233	-0.078	-0.147	0.506
347	1	1.249	-3.10	44.99	-0.224	-0.232	-0.244	-0.080	-0.152	0.517
347	2	1.250	0.04	44.99	-0.225	-0.237	-0.240	-0.081	-0.152	0.529
347	3	1.250	1.10	44.99	-0.233	-0.243	-0.248	-0.084	-0.157	0.541
347	4	1.251	3.20	44.99	-0.242	-0.251	-0.257	-0.087	-0.163	0.562
347	5	1.250	6.35	44.99	-0.232	-0.256	-0.269	-0.083	-0.168	0.565
347	6	1.251	9.52	44.99	-0.254	-0.289	-0.279	-0.091	-0.182	0.591
347	7	1.251	12.65	44.99	-0.273	-0.302	-0.293	-0.098	-0.190	0.613
347	8	1.252	0.07	44.99	-0.223	-0.234	-0.238	-0.080	-0.151	0.527
348	1	1.251	-3.05	44.99	-0.231	-0.236	-0.251	-0.083	-0.156	0.542
348	2	1.251	0.08	44.99	-0.238	-0.240	-0.244	-0.082	-0.155	0.553
348	3	1.250	1.13	44.99	-0.244	-0.251	-0.259	-0.086	-0.160	0.567
348	4	1.251	3.27	44.99	-0.233	-0.255	-0.259	-0.088	-0.164	0.589
348	5	1.250	6.37	44.99	-0.254	-0.291	-0.270	-0.084	-0.167	0.589
348	6	1.250	9.53	44.99	-0.275	-0.291	-0.281	-0.091	-0.183	0.617
348	7	1.250	12.68	44.99	-0.275	-0.307	-0.295	-0.099	-0.193	0.640
348	8	1.251	0.09	44.99	-0.228	-0.241	-0.244	-0.082	-0.155	0.552
349	1	1.252	-2.99	44.99	-0.238	-0.241	-0.257	-0.085	-0.159	0.586
349	2	1.252	1.14	44.99	-0.239	-0.245	-0.248	-0.086	-0.158	0.612
349	3	1.253	3.20	44.99	-0.247	-0.254	-0.250	-0.089	-0.167	0.625
349	4	1.252	6.39	44.99	-0.231	-0.251	-0.263	-0.089	-0.167	0.648
349	5	1.253	9.53	44.99	-0.255	-0.289	-0.268	-0.083	-0.166	0.678
349	6	1.253	12.63	44.99	-0.275	-0.289	-0.279	-0.091	-0.181	0.705

TABLE VA
 BALANCE CAVITY AND BASE PRESSURE COEFFICIENTS, AND INCREMENTAL
 AXIAL FORCE COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	M _c	α	Φ	C _{p_c}	C _{p_{b1}}	C _{p_{b2}}	ΔC _{A_c}	ΔC _{A_b}	C _{A_u}
A49	7	1.252	12.65	44.99	-0.282	-0.307	-0.299	-0.10175	-0.19426	0.70431
A49	11	1.252	0.15	44.99	-0.228	-0.240	-0.243	-0.08231	-0.15487	0.61316
A50	1	1.251	6.35	44.99	-0.229	-0.253	-0.270	-0.08278	-0.16757	0.56600
A50	2	1.249	8.47	44.99	-0.246	-0.282	-0.275	-0.08861	-0.17861	0.58297
A51	1	1.249	6.35	44.99	-0.234	-0.255	-0.275	-0.08453	-0.16990	0.56719
A51	2	1.251	6.35	44.99	-0.235	-0.255	-0.274	-0.08490	-0.16930	0.56930
A52	1	1.247	4.23	44.99	-0.243	-0.252	-0.264	-0.08757	-0.16552	0.57146
A53	1	0.899	-0.04	90.00	-0.113	-0.126	-0.134	-0.04072	-0.08373	0.28535
A53	2	0.899	-0.04	78.80	-0.114	-0.128	-0.135	-0.04135	-0.08561	0.28621
A53	3	0.900	-0.04	67.50	-0.113	-0.126	-0.130	-0.04087	-0.08402	0.28670
A53	4	0.900	-0.04	56.30	-0.111	-0.126	-0.130	-0.04003	-0.08235	0.28664
A53	5	0.899	-0.04	45.00	-0.114	-0.131	-0.136	-0.04136	-0.08579	0.28654
A53	6	0.900	-0.04	33.80	-0.117	-0.128	-0.132	-0.04026	-0.08349	0.28664
A53	7	0.899	-0.04	22.50	-0.113	-0.129	-0.137	-0.04224	-0.08649	0.28661
A53	8	0.899	-0.04	11.30	-0.114	-0.126	-0.134	-0.04098	-0.08458	0.28650
A53	9	0.900	-0.04	0.00	-0.114	-0.126	-0.135	-0.04129	-0.08442	0.28662
A53	10	0.900	-0.04	11.29	-0.114	-0.126	-0.131	-0.04123	-0.08303	0.28664
A53	11	0.899	-0.04	22.49	-0.114	-0.126	-0.131	-0.04155	-0.08549	0.28658
A53	12	0.899	-0.04	33.79	-0.114	-0.126	-0.132	-0.04123	-0.08548	0.28659
A53	13	0.899	-0.04	44.99	-0.116	-0.127	-0.135	-0.04155	-0.08549	0.28672
A53	14	0.899	-0.04	56.29	-0.111	-0.127	-0.130	-0.04135	-0.08271	0.28681
A53	15	0.900	-0.04	67.49	-0.114	-0.130	-0.133	-0.04116	-0.08437	0.28682
A53	16	0.899	-0.04	78.79	-0.111	-0.127	-0.130	-0.04131	-0.08268	0.28693
A53	17	0.900	-0.04	89.99	-0.111	-0.127	-0.130	-0.04031	-0.08268	0.28693
A54	3	1.250	3.13	0.00	-0.198	-0.198	-0.223	-0.07155	-0.13531	0.41852
A54	4	1.250	-0.02	0.00	-0.193	-0.205	-0.210	-0.06956	-0.13523	0.40585
A54	5	1.249	1.02	0.00	-0.192	-0.208	-0.211	-0.06917	-0.13446	0.41032
A54	6	1.249	3.13	0.00	-0.201	-0.214	-0.217	-0.07248	-0.13833	0.41667
A54	7	1.250	6.27	0.00	-0.223	-0.235	-0.244	-0.08051	-0.15364	0.44215
A54	8	1.249	9.38	0.00	-0.256	-0.262	-0.271	-0.09231	-0.17076	0.46891
A54	9	1.247	12.57	0.00	-0.286	-0.288	-0.297	-0.10301	-0.18753	0.49191
A55	3	0.896	3.02	0.00	-0.124	-0.114	-0.143	-0.04474	-0.08236	0.22668
A55	4	0.897	-0.03	0.00	-0.118	-0.130	-0.134	-0.04258	-0.08481	0.21961
A55	5	0.897	0.98	0.00	-0.119	-0.133	-0.131	-0.04293	-0.08491	0.22046
A55	6	0.897	3.09	0.00	-0.124	-0.136	-0.142	-0.04483	-0.08933	0.23045
A55	7	0.900	6.16	0.00	-0.142	-0.156	-0.165	-0.05116	-0.10386	0.24722

TABLE VA
BALANCE CAVITY AND BASE PRESSURE COEFFICIENTS, AND INCREMENTAL AXIAL FORCE COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	M _c	α	φ	C _p c	C _p b ₁	C _p b ₂	ΔC _A c	ΔC _A b	C _A u
A55	8	0.900	9.27	-0.00	-0.169	-0.185	-0.190	-0.06119	-0.12029	0.26392
A55	9	0.897	12.43	-0.00	-0.197	-0.207	-0.216	-0.07094	-0.13568	0.27189
A55	10	0.897	-0.01	-0.00	-0.117	-0.128	-0.132	-0.04222	-0.08364	0.21900
A56	1	0.897	-3.05	-0.00	-0.11A	-0.105	-0.148	-0.04251	-0.08153	0.22622
A56	2	0.89A	-0.01	-0.00	-0.124	-0.139	-0.145	-0.04486	-0.09100	0.23462
A56	3	0.897	1.02	-0.00	-0.131	-0.145	-0.151	-0.04731	-0.09510	0.24422
A56	4	0.894	3.13	-0.00	-0.130	-0.136	-0.149	-0.0469A	-0.09216	0.25194
A56	5	0.897	6.19	-0.00	-0.146	-0.160	-0.172	-0.05268	-0.10651	0.26504
A56	6	0.897	9.26	-0.00	-0.167	-0.1A4	-0.191	-0.06034	-0.12032	0.28105
A56	7	0.897	12.41	-0.00	-0.195	-0.206	-0.215	-0.07052	-0.13505	0.28865
A56	8	0.897	-0.01	-0.00	-0.122	-0.137	-0.142	-0.04424	-0.08943	0.23508
B57	1	0.895	-2.99	-0.00	-0.121	-0.114	-0.159	-0.04388	-0.08793	0.26191
A57	2	0.89A	0.02	-0.00	-0.130	-0.152	-0.154	-0.04681	-0.09819	0.27773
A57	3	0.89A	1.08	-0.00	-0.132	-0.153	-0.153	-0.04782	-0.09842	0.28388
A57	4	0.89A	3.14	-0.00	-0.140	-0.151	-0.161	-0.05069	-0.10013	0.28814
A57	5	0.895	6.21	-0.00	-0.149	-0.159	-0.175	-0.05396	-0.10702	0.29984
A57	6	0.899	9.30	-0.00	-0.164	-0.179	-0.184	-0.05919	-0.11643	0.31035
A57	7	0.896	12.45	-0.00	-0.195	-0.209	-0.214	-0.07037	-0.13473	0.31459
A57	8	0.899	0.02	-0.00	-0.128	-0.149	-0.150	-0.04617	-0.09601	0.27652
B58	1	0.890	-2.95	44.99	-0.124	-0.160	-0.169	-0.04499	-0.10565	0.31075
A58	2	0.89A	0.11	44.99	-0.127	-0.146	-0.154	-0.04580	-0.10963	0.32616
A58	3	0.895	1.13	44.99	-0.140	-0.157	-0.155	-0.05047	-0.10013	0.33673
A58	4	0.897	3.17	44.99	-0.146	-0.169	-0.163	-0.05274	-0.10653	0.35046
A58	5	0.896	6.2A	44.99	-0.143	-0.168	-0.1A4	-0.05176	-0.11280	0.35045
A58	6	0.896	9.35	44.99	-0.163	-0.189	-0.217	-0.058A1	-0.11345	0.38090
A58	7	0.896	12.40	44.99	-0.187	-0.226	-0.236	-0.06753	-0.1480A	0.38251
A58	8	0.89A	0.09	44.99	-0.128	-0.148	-0.155	-0.04633	-0.09722	0.32498
B59	1	0.897	-3.03	44.99	-0.120	-0.154	-0.150	-0.04336	-0.09752	0.23397
A59	2	0.897	0.07	44.99	-0.126	-0.141	-0.142	-0.04547	-0.09087	0.24702
A59	3	0.89A	1.14	44.99	-0.143	-0.154	-0.146	-0.05171	-0.09648	0.25945
A59	4	0.897	3.14	44.99	-0.131	-0.150	-0.152	-0.04738	-0.09679	0.26062
A59	5	0.89A	6.20	44.99	-0.144	-0.178	-0.196	-0.05209	-0.12015	0.27621
A59	6	0.901	9.30	44.99	-0.163	-0.202	-0.217	-0.05889	-0.13458	0.29465
A59	7	0.899	12.40	44.99	-0.191	-0.235	-0.243	-0.06889	-0.15325	0.30570
A59	8	0.899	0.04	44.99	-0.129	-0.142	-0.142	-0.04656	-0.09124	0.24392
B60	1	0.89A	-3.10	44.99	-0.119	-0.142	-0.140	-0.04316	-0.09078	0.22391
A60	2	0.897	-0.01	44.99	-0.115	-0.121	-0.127	-0.04141	-0.07995	0.21409

TABLE VA
BALANCE CAVITY AND BASE PRESSURE COEFFICIENTS, AND INCREMENTAL
AXIAL FORCE COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	M _c	α	Φ	C _{Pc}	C _{Pb1}	C _{Pb2}	ΔC _{Ac}	ΔC _{Ab}	C _{Au}
A60	3	0.897	1.02	44.99	-0.119	-0.131	-0.133	0.04294	-0.08485	0.21628
A60	4	0.900	3.08	44.99	-0.124	-0.140	-0.150	0.04473	-0.09332	0.22584
A60	5	0.898	6.18	44.99	-0.145	-0.180	-0.199	0.05221	-0.12139	0.24652
A60	6	0.898	9.28	44.99	-0.166	-0.206	-0.223	0.06005	-0.13760	0.26232
A60	7	0.898	12.38	44.99	-0.192	-0.235	-0.245	0.06916	-0.15385	0.26882
A60	8	0.897	0.01	44.99	-0.118	-0.126	-0.132	0.04274	-0.08281	0.21242
A61	6	0.600	-3.08	-0.00	-0.088	-0.087	-0.107	0.03190	-0.06257	0.21948
A61	7	0.600	-0.03	-0.00	-0.084	-0.107	-0.097	0.03030	-0.06558	0.21190
A61	8	0.600	1.01	-0.00	-0.086	-0.113	-0.097	0.03098	-0.06732	0.21626
A61	9	0.600	3.08	-0.00	-0.090	-0.121	-0.103	0.03271	-0.07175	0.22198
A61	10	0.599	6.19	-0.00	-0.113	-0.152	-0.130	0.04072	-0.09068	0.23805
A61	11	0.600	9.26	-0.00	-0.134	-0.164	-0.156	0.04826	-0.10259	0.25556
A61	12	0.599	12.41	-0.00	-0.159	-0.173	-0.175	0.05741	-0.11179	0.26418
A61	13	0.600	-0.03	-0.00	-0.079	-0.102	-0.091	0.02852	-0.06196	0.21149
A62	1	0.599	-3.03	-0.00	-0.089	-0.087	-0.114	0.03235	-0.06452	0.22208
A62	2	0.600	0.00	-0.00	-0.085	-0.110	-0.104	0.03085	-0.06393	0.22528
A62	3	0.600	1.03	-0.00	-0.090	-0.112	-0.106	0.03246	-0.07033	0.23028
A62	4	0.599	3.14	-0.00	-0.092	-0.113	-0.107	0.03313	-0.07074	0.23355
A62	5	0.601	6.18	-0.00	-0.107	-0.145	-0.126	0.03860	-0.08706	0.25707
A62	6	0.599	9.27	-0.00	-0.131	-0.162	-0.152	0.04743	-0.10084	0.27073
A62	7	0.600	12.40	-0.00	-0.156	-0.175	-0.173	0.05616	-0.11105	0.28000
A62	8	0.600	-0.02	-0.00	-0.088	-0.111	-0.106	0.03170	-0.06995	0.22591
A63	1	0.600	-2.97	-0.00	-0.092	-0.092	-0.121	0.03328	-0.06863	0.25399
A63	2	0.600	0.04	-0.00	-0.092	-0.122	-0.115	0.03321	-0.07623	0.26491
A63	3	0.600	1.05	-0.00	-0.093	-0.126	-0.114	0.03367	-0.07709	0.26869
A63	4	0.598	3.16	-0.00	-0.098	-0.121	-0.113	0.03537	-0.07532	0.27512
A63	5	0.599	6.21	-0.00	-0.108	-0.144	-0.128	0.03923	-0.08712	0.28531
A63	6	0.600	9.31	-0.00	-0.130	-0.167	-0.148	0.04681	-0.10126	0.29570
A63	7	0.600	12.46	-0.00	-0.149	-0.173	-0.166	0.05393	-0.10888	0.29825
A63	8	0.601	0.03	-0.00	-0.088	-0.122	-0.114	0.03201	-0.07587	0.26243
A63	9	0.601	-2.96	44.99	-0.092	-0.138	-0.125	0.03345	-0.08458	0.29413
A63	10	0.599	0.11	44.99	-0.107	-0.126	-0.117	0.03641	-0.07866	0.32205
A63	11	0.600	3.16	44.99	-0.107	-0.136	-0.118	0.03868	-0.08134	0.32035
A63	12	0.600	6.24	44.99	-0.095	-0.135	-0.118	0.03443	-0.08106	0.32291
A63	13	0.601	-0.02	44.99	-0.107	-0.147	-0.150	0.03866	-0.09529	0.33291
A64	1	0.600	-2.99	44.99	-0.095	-0.140	-0.126	0.03421	-0.08547	0.29452
A64	2	0.600	0.10	44.99	-0.094	-0.124	-0.115	0.03405	-0.07682	0.30758
A64	3	0.601	1.11	44.99	-0.102	-0.132	-0.112	0.03695	-0.07838	0.32007

TABLE VA
BALANCE CAVITY AND BASE PRESSURE COEFFICIENTS, AND INCREMENTAL
AXIAL FORCE COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	M _c	α	Φ	C _{Pc}	C _{Pb1}	C _{Pb2}	ΔC _{Ac}	ΔC _{Ab}	C _{Au}
A64	0.600	3.17	44.99	-0.097	-0.135	-0.124	-0.03493	-0.06322	0.32058
A64	0.600	6.23	44.99	-0.106	-0.147	-0.147	-0.03851	-0.09422	0.33246
A64	0.599	9.31	44.99	-0.133	-0.168	-0.192	-0.04810	-0.12215	0.34575
A64	0.600	12.39	44.99	-0.148	-0.207	-0.208	-0.05333	-0.13330	0.34693
A64	0.600	0.0A	44.99	-0.095	-0.123	-0.110	-0.03426	-0.07498	0.30769
A65	0.599	-3.06	44.99	-0.092	-0.136	-0.118	-0.03334	-0.08154	0.23091
A65	0.599	0.03	44.99	-0.094	-0.116	-0.104	-0.03410	-0.07081	0.23629
A65	0.600	1.07	44.99	-0.095	-0.116	-0.101	-0.03426	-0.06998	0.24147
A65	0.599	3.12	44.99	-0.096	-0.125	-0.111	-0.03466	-0.07567	0.24926
A65	0.600	6.20	44.99	-0.110	-0.154	-0.159	-0.03983	-0.10030	0.26762
A65	0.600	9.30	44.99	-0.128	-0.186	-0.184	-0.04642	-0.11842	0.28500
A65	0.600	12.38	44.99	-0.152	-0.214	-0.209	-0.05476	-0.13557	0.29915
A65	0.599	0.01	44.99	-0.092	-0.116	-0.104	-0.03347	-0.07079	0.23355
A66	0.600	-3.12	45.00	-0.087	-0.121	-0.104	-0.03151	-0.07242	0.22173
A66	0.600	0.03	44.99	-0.083	-0.106	-0.096	-0.02999	-0.06419	0.21501
A66	0.601	3.05	44.99	-0.081	-0.106	-0.097	-0.02918	-0.06527	0.21598
A66	0.601	6.16	44.99	-0.092	-0.123	-0.115	-0.03332	-0.07641	0.22442
A66	0.601	9.26	44.99	-0.111	-0.158	-0.164	-0.04013	-0.10335	0.24340
A66	0.600	12.35	44.99	-0.133	-0.189	-0.190	-0.04792	-0.12144	0.25749
A66	0.600	0.04	44.99	-0.082	-0.103	-0.094	-0.02959	-0.06345	0.21346
A67	1.250	-3.12	-0.00	-0.215	-0.213	-0.239	-0.07766	-0.14500	0.42530
A67	1.250	1.03	-0.00	-0.207	-0.221	-0.227	-0.07484	-0.14363	0.41287
A67	1.250	3.14	0.00	-0.211	-0.226	-0.231	-0.07601	-0.14660	0.41777
A67	1.251	6.27	0.00	-0.241	-0.251	-0.240	-0.07853	-0.15150	0.42521
A67	1.251	9.40	0.00	-0.274	-0.276	-0.265	-0.08694	-0.16531	0.45187
A67	1.252	12.59	0.00	-0.300	-0.303	-0.294	-0.09875	-0.18278	0.47857
A67	1.251	0.03	-0.00	-0.209	-0.222	-0.228	-0.07552	-0.14423	0.41997
A68	1.252	-3.06	0.00	-0.213	-0.217	-0.238	-0.07669	-0.14591	0.42840
A68	1.250	1.03	-0.00	-0.218	-0.232	-0.239	-0.07862	-0.15096	0.43818
A68	1.249	3.20	0.00	-0.223	-0.235	-0.247	-0.08062	-0.15488	0.44600
A68	1.250	6.30	0.00	-0.225	-0.235	-0.249	-0.08134	-0.15523	0.45163
A68	1.250	9.46	0.00	-0.242	-0.250	-0.268	-0.08714	-0.16613	0.47246
A68	1.249	12.65	0.00	-0.272	-0.273	-0.295	-0.09797	-0.18201	0.50157
A68	1.250	0.01	-0.00	-0.303	-0.298	-0.317	-0.10925	-0.19743	0.52730
A68	1.250	0.01	-0.00	-0.226	-0.240	-0.248	-0.08144	-0.15625	0.43553

TABLE VA
BALANCE CAVITY AND BASE PRESSURE COEFFICIENTS, AND INCREMENTAL
AXIAL FORCE COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	M _c	α	φ	C _p c	C _p b ₁	C _p b ₂	ΔC _A c	ΔC _A b	C _A u
A69	1	1.249	-2.99	-0.00	-0.228	-0.231	-0.252	-0.08231	-0.15482	0.46468
A69	2	1.249	-0.10	-0.00	-0.233	-0.247	-0.253	-0.08410	-0.16016	0.46312
A69	3	1.251	1.13	-0.00	-0.234	-0.248	-0.261	-0.08450	-0.16318	0.49470
A69	4	1.249	3.24	-0.00	-0.238	-0.246	-0.262	-0.08575	-0.16294	0.50163
A69	5	1.250	6.35	-0.00	-0.246	-0.253	-0.273	-0.08859	-0.16865	0.51603
A69	6	1.249	9.47	-0.00	-0.270	-0.274	-0.296	-0.09750	-0.18272	0.54339
A69	7	1.249	12.68	0.00	-0.301	-0.296	-0.318	-0.10852	-0.19686	0.56910
A69	8	1.250	0.10	-0.00	-0.232	-0.245	-0.251	-0.08378	-0.15886	0.47907
A70	1	1.249	-2.95	44.99	-0.237	-0.252	-0.267	-0.08556	-0.16617	0.51561
A70	2	1.249	-0.16	44.99	-0.245	-0.260	-0.255	-0.08828	-0.16522	0.54246
A70	3	1.250	1.20	44.99	-0.250	-0.271	-0.257	-0.09031	-0.16916	0.55453
A70	4	1.249	3.26	44.99	-0.255	-0.267	-0.265	-0.09196	-0.17063	0.56811
A70	5	1.250	6.36	44.99	-0.258	-0.258	-0.276	-0.08726	-0.17123	0.57132
A70	6	1.249	9.51	44.99	-0.272	-0.257	-0.319	-0.09805	-0.20068	0.60859
A70	7	1.251	12.62	44.99	-0.298	-0.330	-0.341	-0.10742	-0.21489	0.63126
A70	8	1.249	0.17	44.99	-0.246	-0.260	-0.255	-0.08858	-0.16533	0.53992
A71	1	1.249	-3.07	44.99	-0.229	-0.247	-0.253	-0.08276	-0.16032	0.43777
A71	2	1.251	-0.06	44.99	-0.235	-0.248	-0.244	-0.08469	-0.15776	0.45289
A71	3	1.250	1.19	44.99	-0.245	-0.258	-0.252	-0.08833	-0.16365	0.46297
A71	4	1.250	3.32	44.99	-0.248	-0.257	-0.261	-0.08988	-0.16522	0.47818
A71	5	1.250	6.47	44.99	-0.274	-0.267	-0.286	-0.08929	-0.17713	0.48796
A71	6	1.250	9.59	44.99	-0.299	-0.307	-0.320	-0.09892	-0.20103	0.52149
A71	7	1.250	12.59	44.99	-0.323	-0.321	-0.341	-0.10768	-0.21223	0.54400
A71	8	1.250	0.07	44.99	-0.237	-0.249	-0.245	-0.08550	-0.15853	0.45091
A72	1	1.250	-3.14	44.99	-0.231	-0.239	-0.248	-0.08347	-0.15627	0.43075
A72	2	1.251	-0.02	44.99	-0.217	-0.224	-0.233	-0.07824	-0.14634	0.41108
A72	3	1.251	1.02	44.99	-0.218	-0.227	-0.234	-0.07868	-0.14779	0.41511
A72	4	1.250	3.12	44.99	-0.233	-0.227	-0.252	-0.08400	-0.15799	0.42996
A72	5	1.251	6.24	44.99	-0.253	-0.278	-0.292	-0.09133	-0.18287	0.45770
A72	6	1.250	9.40	44.99	-0.283	-0.312	-0.323	-0.10207	-0.20332	0.48409
A72	7	1.251	12.54	44.99	-0.302	-0.326	-0.343	-0.10875	-0.21454	0.49942
A72	8	1.251	-0.00	44.99	-0.216	-0.223	-0.231	-0.07782	-0.14556	0.40983
A73	3	1.250	-2.99	0.00	-0.221	-0.226	-0.246	-0.07962	-0.15142	0.48575
A73	4	1.250	1.10	-0.00	-0.227	-0.244	-0.248	-0.08184	-0.15780	0.50697
A73	5	1.250	3.25	-0.00	-0.234	-0.251	-0.261	-0.08453	-0.16401	0.51602
A73	6	1.250	6.36	0.00	-0.237	-0.246	-0.262	-0.08537	-0.16275	0.52522
A73	7	1.251	9.48	0.00	-0.242	-0.253	-0.273	-0.08719	-0.16858	0.54379
A73	8	1.249	12.65	0.00	-0.271	-0.276	-0.299	-0.09775	-0.18454	0.57280
A73	9	1.250	0.00	0.00	-0.299	-0.297	-0.319	-0.10776	-0.19754	0.59823

TABLE VA
BALANCE CAVITY AND BASE PRESSURE COEFFICIENTS, AND INCREMENTAL AXIAL FORCE COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	M _c	α	Φ	C _p c	C _p b ₁	C _p b ₂	ΔC _A c	ΔC _A b	C _A u
873	10	1.251	0.06	-0.00	-0.230	-0.246	-0.250	-0.08290	-0.15911	0.50456
874	1	1.250	-3.07	0.00	-0.221	-0.241	-0.249	-0.07979	-0.15072	0.45038
874	2	1.250	0.03	0.00	-0.223	-0.240	-0.248	-0.08053	-0.15665	0.46078
874	3	1.250	1.08	0.00	-0.229	-0.246	-0.255	-0.08271	-0.16068	0.46894
874	4	1.248	3.18	0.00	-0.230	-0.242	-0.256	-0.08311	-0.15969	0.47550
874	5	1.247	6.33	0.00	-0.247	-0.258	-0.276	-0.08908	-0.17116	0.49851
874	6	1.250	9.42	0.00	-0.275	-0.278	-0.301	-0.09912	-0.18562	0.52923
874	7	1.251	12.65	0.00	-0.302	-0.299	-0.322	-0.10882	-0.19899	0.55523
874	8	1.249	0.02	0.00	-0.224	-0.240	-0.248	-0.08072	-0.15631	0.45739
875	1	1.250	-3.13	0.00	-0.221	-0.217	-0.244	-0.07989	-0.14785	0.44877
875	2	1.250	-0.02	0.00	-0.209	-0.224	-0.230	-0.07552	-0.14551	0.43727
875	3	1.249	1.04	0.00	-0.215	-0.229	-0.237	-0.07741	-0.14966	0.44283
875	4	1.249	3.12	0.00	-0.225	-0.239	-0.249	-0.08119	-0.15658	0.47858
875	5	1.249	6.27	0.00	-0.250	-0.259	-0.276	-0.09004	-0.17155	0.50869
875	6	1.250	9.41	0.00	-0.282	-0.286	-0.305	-0.10152	-0.18934	0.54869
875	7	1.249	12.62	0.00	-0.309	-0.311	-0.328	-0.11136	-0.20484	0.58293
875	8	1.248	-0.04	0.00	-0.212	-0.226	-0.232	-0.07647	-0.14679	0.43642
876	1	0.900	-3.08	0.00	-0.124	-0.113	-0.144	-0.04478	-0.08273	0.3038
876	2	0.899	-0.04	0.00	-0.117	-0.131	-0.132	-0.04238	-0.08441	0.22315
876	3	0.899	1.00	0.00	-0.121	-0.138	-0.132	-0.04363	-0.08497	0.22339
876	4	0.898	3.10	0.00	-0.129	-0.146	-0.150	-0.04656	-0.09497	0.23338
876	5	0.898	6.19	0.00	-0.148	-0.166	-0.173	-0.05357	-0.10886	0.25396
876	6	0.898	9.26	0.00	-0.176	-0.190	-0.197	-0.06350	-0.12424	0.26877
876	7	0.897	12.41	0.00	-0.203	-0.213	-0.222	-0.07322	-0.13940	0.27744
876	8	0.900	-0.04	0.00	-0.115	-0.129	-0.131	-0.04164	-0.08356	0.22315
877	1	0.898	-3.05	0.00	-0.125	-0.114	-0.155	-0.04506	-0.08652	0.23259
877	2	0.897	-0.05	0.00	-0.132	-0.145	-0.152	-0.04767	-0.09562	0.24349
877	3	0.899	1.05	0.00	-0.134	-0.147	-0.152	-0.04761	-0.09616	0.25329
877	4	0.898	3.14	0.00	-0.153	-0.161	-0.177	-0.05517	-0.10970	0.27177
877	5	0.896	6.23	0.00	-0.168	-0.186	-0.193	-0.06068	-0.12141	0.28670
877	6	0.897	9.30	0.00	-0.196	-0.216	-0.224	-0.07439	-0.14127	0.29658
877	7	0.895	12.43	0.00	-0.206	-0.216	-0.222	-0.07439	-0.14127	0.29658
877	8	0.897	0.00	0.00	-0.131	-0.145	-0.151	-0.04739	-0.09505	0.23931
878	1	0.898	-2.97	0.00	-0.126	-0.117	-0.163	-0.04539	-0.09004	0.26782
878	2	0.897	-0.06	0.00	-0.135	-0.156	-0.159	-0.04878	-0.10109	0.28256
878	3	0.899	1.08	0.00	-0.138	-0.158	-0.159	-0.04983	-0.10173	0.28990
878	4	0.901	3.15	0.00	-0.137	-0.148	-0.159	-0.04949	-0.09843	0.29632

TABLE VA
BALANCE CAVITY AND BASE PRESSURE COEFFICIENTS, AND INCREMENTAL AXIAL FORCE COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	M _c	α	Φ	C _{Pc}	C _{Pbl}	C _{Pb2}	ΔC _{Ac}	ΔC _{Ab}	C _{Au}
878	0.899	6.23	-0.00	-0.151	-0.161	-0.177	-0.05459	-0.10827	0.30726
A7A	0.89A	9.31	0.00	-0.176	-0.193	-0.199	-0.06366	-0.12570	0.31742
A7A	0.89A	12.44	0.00	-0.199	-0.208	-0.218	-0.07182	-0.13645	0.32411
A7A	0.900	0.05	-0.00	-0.130	-0.149	-0.152	-0.04709	-0.09682	0.28250
A79	0.59A	-2.97	-0.00	-0.104	-0.104	-0.131	-0.03753	-0.07574	0.26602
A79	0.59A	0.01	-0.00	-0.106	-0.132	-0.130	-0.03824	-0.08519	0.27681
A79	0.59A	1.08	-0.00	-0.109	-0.138	-0.130	-0.03948	-0.08616	0.28119
A79	0.59A	3.12	-0.00	-0.110	-0.134	-0.127	-0.03989	-0.08399	0.28495
A79	0.59A	6.24	-0.00	-0.120	-0.149	-0.138	-0.04347	-0.09210	0.29775
A79	0.59A	9.25	-0.00	-0.141	-0.173	-0.160	-0.05092	-0.10679	0.30737
A79	0.599	12.44	-0.00	-0.165	-0.1A7	-0.183	-0.05961	-0.11870	0.31220
A79	0.599	0.07	-0.00	-0.101	-0.131	-0.122	-0.03661	-0.08138	0.27483
A80	0.59A	-3.04	-0.00	-0.103	-0.104	-0.125	-0.03742	-0.07368	0.23631
A80	0.59A	0.04	-0.00	-0.103	-0.127	-0.116	-0.03608	-0.07639	0.23773
A80	0.59A	1.04	-0.00	-0.103	-0.127	-0.117	-0.03737	-0.07841	0.24185
A80	0.599	3.15	-0.00	-0.106	-0.128	-0.120	-0.03819	-0.07972	0.25057
A80	0.59A	6.21	-0.00	-0.122	-0.156	-0.137	-0.04401	-0.09415	0.26774
A80	0.59A	9.30	-0.00	-0.145	-0.176	-0.162	-0.05230	-0.10855	0.28432
A80	0.59A	12.41	-0.00	-0.173	-0.189	-0.192	-0.06244	-0.12222	0.29479
A80	0.597	-0.00	0.00	-0.097	-0.117	-0.114	-0.03509	-0.07446	0.23889
A81	0.597	-3.10	-0.00	-0.104	-0.104	-0.121	-0.03774	-0.07221	0.23292
A81	0.59A	0.01	-0.00	-0.09A	-0.118	-0.106	-0.03546	-0.07211	0.22844
A81	0.59A	0.99	-0.00	-0.096	-0.119	-0.103	-0.03472	-0.07130	0.22950
A81	0.59A	3.11	-0.00	-0.106	-0.134	-0.115	-0.03816	-0.07992	0.23478
A81	0.598	6.17	-0.00	-0.121	-0.161	-0.133	-0.04387	-0.09413	0.24941
A81	0.597	9.28	-0.00	-0.150	-0.179	-0.166	-0.05410	-0.11085	0.27007
A81	0.59A	12.40	-0.00	-0.175	-0.191	-0.192	-0.06316	-0.12275	0.27934
A81	0.59A	-0.06	-0.00	-0.094	-0.112	-0.101	-0.03419	-0.06856	0.22734
A83	1.251	-3.07	0.00	-0.217	-0.214	-0.242	-0.07813	-0.14637	0.41660
A83	1.249	-1.05	0.00	-0.213	-0.226	-0.231	-0.07690	-0.14681	0.41109
A83	1.250	0.03	0.00	-0.210	-0.225	-0.228	-0.07562	-0.14515	0.40930
A83	1.249	0.49	0.00	-0.215	-0.228	-0.232	-0.07746	-0.14747	0.41090
A83	1.250	1.01	0.00	-0.214	-0.224	-0.231	-0.07696	-0.14707	0.41412
A83	1.251	3.14	0.00	-0.257	-0.286	-0.278	-0.09275	-0.18080	0.45640
A83	1.250	6.14	0.00	-0.28A	-0.312	-0.307	-0.10391	-0.19838	0.47933
A83	1.250	9.24	0.00	-0.312	-0.321	-0.326	-0.11264	-0.20745	0.49248
A83	1.251	-0.05	0.00	-0.211	-0.225	-0.229	-0.07597	-0.14565	0.40888

TABLE VA
 BALANCE CAVITY AND BASE PRESSURE COEFFICIENTS, AND INCREMENTAL
 AXIAL FORCE COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	M _c	α	Φ	C _p c	C _p b1	C _p b2	ΔC _A c	ΔC _A b	C _A u
A84	1	1.046	-3.06	0.00	-0.227	-0.217	-0.260	-0.08197	-0.15301	0.41530
A84	2	1.046	-1.07	0.00	-0.222	-0.229	-0.241	-0.08000	-0.15081	0.41487
A84	3	1.047	-0.03	0.00	-0.225	-0.239	-0.240	-0.08114	-0.15366	0.40249
A84	4	1.048	0.48	0.00	-0.219	-0.233	-0.233	-0.07903	-0.15034	0.40293
A84	5	1.047	0.99	0.00	-0.216	-0.233	-0.237	-0.07786	-0.14806	0.41866
A84	6	1.050	3.15	0.00	-0.225	-0.252	-0.278	-0.08118	-0.15694	0.41396
A84	7	1.047	6.15	0.00	-0.236	-0.293	-0.306	-0.09233	-0.18313	0.47485
A84	8	1.047	9.22	0.00	-0.233	-0.322	-0.325	-0.10577	-0.20144	0.47089
A84	9	1.047	12.36	0.00	-0.318	-0.329	-0.325	-0.11482	-0.20942	0.44054
A84	10	1.047	-0.03	0.00	-0.221	-0.235	-0.235	-0.07984	-0.15060	0.44054
A85	1	0.99A	3.06	0.00	-0.212	-0.206	-0.249	-0.07639	-0.14571	0.34698
A85	2	1.002	-1.03	0.00	-0.206	-0.210	-0.225	-0.07428	-0.13964	0.34703
A85	3	1.002	-0.04	0.00	-0.209	-0.216	-0.221	-0.07524	-0.14094	0.33722
A85	4	1.001	1.48	0.00	-0.204	-0.217	-0.216	-0.07362	-0.13902	0.34414
A85	5	1.001	3.05	0.00	-0.219	-0.223	-0.218	-0.07542	-0.14146	0.34419
A85	6	1.002	6.12	0.00	-0.218	-0.245	-0.226	-0.07854	-0.15094	0.35563
A85	7	1.002	9.18	0.00	-0.254	-0.295	-0.270	-0.09171	-0.18118	0.39254
A85	8	1.002	12.35	0.00	-0.297	-0.325	-0.307	-0.10705	-0.20284	0.43360
A85	9	0.999	-0.03	0.00	-0.205	-0.216	-0.231	-0.11761	-0.21284	0.43360
A85	10	1.001	-0.03	0.00	-0.205	-0.216	-0.218	-0.07380	-0.13915	0.33934
A86	2	0.947	-1.06	0.00	-0.142	-0.141	-0.159	-0.05067	-0.09623	0.23737
A86	3	0.949	0.49	0.00	-0.144	-0.153	-0.154	-0.05133	-0.09765	0.23774
A86	4	0.947	0.99	0.00	-0.141	-0.154	-0.150	-0.05185	-0.09742	0.23774
A86	5	0.947	3.04	0.00	-0.144	-0.167	-0.154	-0.05210	-0.10296	0.24580
A86	6	0.946	6.12	0.00	-0.142	-0.198	-0.196	-0.06566	-0.13292	0.27073
A86	7	0.947	9.21	0.00	-0.211	-0.239	-0.219	-0.07626	-0.14692	0.29553
A86	8	0.949	12.32	0.00	-0.240	-0.255	-0.244	-0.08649	-0.15992	0.30367
A86	9	0.945	-0.00	0.00	-0.143	-0.152	-0.157	-0.05172	-0.09916	0.23337
A87	1	0.897	3.04	0.00	-0.130	-0.112	-0.155	-0.04687	-0.08572	0.22132
A87	2	0.897	-1.05	0.00	-0.126	-0.127	-0.143	-0.04551	-0.08678	0.21878
A87	3	0.897	0.50	0.00	-0.126	-0.135	-0.137	-0.04565	-0.08728	0.21655
A87	4	0.898	0.99	0.00	-0.125	-0.136	-0.133	-0.04527	-0.08622	0.21673
A87	5	0.898	3.07	0.00	-0.133	-0.141	-0.135	-0.04598	-0.08789	0.21774
A87	6	0.899	6.06	0.00	-0.135	-0.154	-0.144	-0.04798	-0.09572	0.22407
A87	7	0.897	9.16	0.00	-0.166	-0.191	-0.180	-0.05999	-0.11890	0.25078
A87	8	0.897	12.32	0.00	-0.198	-0.208	-0.198	-0.06795	-0.13022	0.26892
A87	9	0.897	-0.00	0.00	-0.115	-0.122	-0.122	-0.07764	-0.14371	0.27356
A87	10	0.899	-0.00	0.00	-0.124	-0.132	-0.134	-0.04475	-0.08561	0.21691

TABLE VA
BALANCE CAVITY AND BASE PRESSURE COEFFICIENTS, AND INCREMENTAL
AXIAL FORCE COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	M _c	α	Φ	C _p c	C _p b1	C _p b2	ΔC _A c	ΔC _A b	C _A u
88A	1	0.799	-3.05	0.00	-0.120	-0.103	-0.142	-0.04320	-0.07070	0.22131
88A	2	0.79A	-1.04	0.00	-0.116	-0.117	-0.131	-0.04187	-0.07951	0.21813
88A	3	0.79A	-0.03	0.00	-0.113	-0.127	-0.125	-0.04101	-0.08109	0.21534
88A	4	0.79A	0.50	0.00	-0.114	-0.131	-0.124	-0.0412A	-0.08190	0.21594
88A	5	0.79A	1.00	0.00	-0.115	-0.134	-0.122	-0.04154	-0.08194	0.21711
88A	6	0.79A	3.06	0.00	-0.121	-0.145	-0.130	-0.04364	-0.08806	0.22325
88A	7	0.797	6.12	0.00	-0.151	-0.175	-0.167	-0.05462	-0.10984	0.24636
88A	8	0.79A	9.21	0.00	-0.179	-0.199	-0.192	-0.06467	-0.12545	0.26649
88A	9	0.796	12.34	0.00	-0.200	-0.212	-0.212	-0.07204	-0.13618	0.27131
88A	10	0.798	-0.03	0.00	-0.116	-0.130	-0.125	-0.04183	-0.08199	0.21472
889	1	0.599	3.06	0.00	-0.106	-0.097	-0.121	-0.03848	-0.07020	0.22115
889	2	0.59A	-1.05	0.00	-0.099	-0.102	-0.107	-0.03596	-0.06721	0.21874
889	3	0.59A	-0.03	0.00	-0.101	-0.115	-0.107	-0.03650	-0.07134	0.21670
889	4	0.59A	0.4A	0.00	-0.099	-0.119	-0.103	-0.03599	-0.07141	0.21760
889	5	0.597	1.02	0.00	-0.101	-0.121	-0.109	-0.03639	-0.07096	0.21755
889	6	0.597	3.08	0.00	-0.108	-0.133	-0.108	-0.03897	-0.07718	0.22470
889	7	0.59A	6.12	0.00	-0.134	-0.165	-0.140	-0.04833	-0.09793	0.24876
889	8	0.597	9.15	0.00	-0.161	-0.179	-0.167	-0.05814	-0.11083	0.26541
889	9	0.597	12.33	0.00	-0.184	-0.194	-0.190	-0.06658	-0.12324	0.27425
889	10	0.597	-0.04	0.00	-0.099	-0.115	-0.105	-0.03574	-0.07070	0.21688

TABLE VB
BALANCE CAVITY AND MODEL BASE PRESSURE COEFFICIENTS AND INCREMENTAL AXIAL FORCE COEFFICIENTS - THRUST EFFECT PHASE

Run	Pt.	M _c	α	Φ	C _{Pc}	C _{Pb1}	C _{Pb2}	ΔC _{Ac}	ΔC _{Ab}	C _{Au}
110	3	1.000	0.00	-0.00	0.863	0.862	0.857	0.07038	0.12781	0.35835
110	4	0.999	-0.00	-0.00	1.366	1.328	1.332	0.18841	0.30247	0.35445
110	5	0.996	-0.00	-0.00	1.442	1.414	1.413	0.22957	0.38136	0.48532
110	6	0.997	-0.00	-0.00	1.485	1.448	1.452	0.24851	0.40970	0.53163
110	7	1.000	-0.01	-0.00	1.506	1.450	1.463	0.26156	0.41991	0.58318
110	8	0.997	-0.00	-0.00	1.522	1.468	1.479	0.26842	0.43255	0.59640
111	5	0.794	-4.03	-0.00	0.943	0.942	0.942	0.04583	0.08350	0.27193
111	6	0.800	-3.03	-0.00	0.942	0.943	0.942	0.04458	0.08131	0.26947
111	7	0.796	-2.08	-0.00	0.943	0.943	0.940	0.04508	0.08277	0.27106
111	8	0.800	-1.01	-0.00	0.943	0.942	0.940	0.04502	0.08246	0.26984
111	9	0.800	-0.54	-0.00	0.943	0.942	0.940	0.04505	0.08240	0.27404
111	10	0.797	-0.00	-0.00	0.944	0.944	0.940	0.04506	0.08230	0.27136
111	11	0.798	0.46	-0.00	0.944	0.944	0.941	0.04438	0.08147	0.26976
111	12	0.799	0.98	-0.00	0.943	0.944	0.939	0.04515	0.08262	0.27255
111	13	0.801	2.02	-0.00	0.943	0.947	0.939	0.04497	0.08022	0.27429
111	14	0.800	3.05	-0.00	0.943	0.950	0.938	0.04548	0.07868	0.27329
111	15	0.801	4.08	-0.00	0.944	0.944	0.941	0.04450	0.08105	0.27011
111	16	0.800	-0.04	-0.00	0.944	0.944	0.941	0.04450	0.08105	0.27011
112	1	0.798	-4.02	-0.00	1.247	1.175	1.186	0.19966	0.26007	0.38254
112	2	0.801	-3.05	-0.00	1.250	1.185	1.197	0.20050	0.27256	0.39057
112	3	0.796	-2.05	-0.00	1.244	1.197	1.203	0.19888	0.28965	0.43010
112	4	0.802	-1.00	-0.00	1.231	1.205	1.216	0.18591	0.30603	0.47965
112	5	0.802	-0.54	-0.00	1.234	1.211	1.216	0.18840	0.30068	0.53719
112	6	0.807	-0.06	-0.00	1.219	1.187	1.186	0.17732	0.26046	0.33483
112	7	0.799	0.47	-0.00	1.225	1.186	1.177	0.17401	0.23572	0.32466
112	8	0.803	0.97	-0.00	1.220	1.170	1.160	0.16559	0.22128	0.31091
112	9	0.803	2.01	-0.00	1.232	1.167	1.148	0.15528	0.22128	0.30089
112	10	0.803	3.04	-0.00	1.239	1.147	1.132	0.15251	0.22985	0.30345
112	11	0.801	-0.03	-0.00	1.207	1.180	1.178	0.16666	0.25495	0.30345
113	4	1.250	-4.09	-0.00	0.770	0.762	0.747	0.07558	0.14313	0.45893
113	5	1.250	-3.10	-0.00	0.768	0.762	0.751	0.07617	0.14224	0.45819
113	6	1.250	-2.10	-0.00	0.768	0.765	0.754	0.07617	0.14030	0.45725
113	7	1.250	-1.05	-0.00	0.768	0.766	0.753	0.07640	0.14031	0.45676
113	8	1.250	-0.56	-0.00	0.768	0.767	0.755	0.07621	0.13853	0.45676
113	9	1.249	-0.07	-0.00	0.768	0.773	0.755	0.07635	0.13773	0.45512
113	10	1.249	0.47	-0.00	0.768	0.777	0.756	0.07634	0.13641	0.45572
113	11	1.249	1.01	-0.00	0.768	0.777	0.756	0.07634	0.13641	0.45572

TABLE VB
BALANCE CAVITY AND MODEL BASE PRESSURE COEFFICIENTS AND INCREMENTAL AXIAL FORCE COEFFICIENTS - THRUST EFFECT PHASE

Run	Pt.	M _c	α	Φ	C _{Pc}	C _{Pb1}	C _{Pb2}	ΔC _{Ac}	ΔC _{Ab}	C _{Au}
113	12	1.249	2.05	-0.00	0.771	0.786	0.759	-0.07520	-0.13226	0.45666
113	13	1.250	3.09	-0.00	0.772	0.794	0.759	-0.07491	-0.13057	0.45650
113	14	1.250	4.11	-0.00	0.771	0.797	0.756	-0.07516	-0.13061	0.45992
113	15	1.251	-0.03	-0.00	0.767	0.771	0.756	-0.07626	-0.13777	0.45447
114	1	1.097	-4.07	-0.00	0.841	0.839	0.829	-0.06766	-0.12566	0.43717
114	2	1.098	-3.07	-0.00	0.847	0.845	0.836	-0.06516	-0.12061	0.42658
114	3	1.096	-2.08	-0.00	0.852	0.853	0.843	-0.06293	-0.11526	0.42087
114	4	1.099	-1.05	-0.00	0.854	0.854	0.845	-0.06204	-0.11370	0.41600
114	5	1.099	-0.55	-0.00	0.852	0.852	0.843	-0.06288	-0.11491	0.41231
114	9	1.097	-0.58	-0.00	0.858	0.856	0.849	-0.06067	-0.11095	0.41316
114	10	1.096	-0.45	-0.00	0.858	0.857	0.850	-0.06038	-0.11154	0.41648
114	11	1.102	0.99	-0.00	0.855	0.856	0.843	-0.06332	-0.11411	0.42048
114	12	1.096	2.03	-0.00	0.853	0.857	0.844	-0.06258	-0.11328	0.42601
114	13	1.099	3.09	-0.00	0.842	0.847	0.834	-0.06708	-0.12023	0.43350
114	14	1.098	4.11	-0.00	0.839	0.847	0.830	-0.06835	-0.12204	0.44036
114	15	1.100	-0.03	-0.00	0.851	0.851	0.843	-0.06322	-0.11504	0.41736
115	1	0.999	-4.08	-0.00	0.843	0.840	0.834	-0.06051	-0.14851	0.42297
115	2	0.998	-3.05	-0.00	0.851	0.850	0.845	-0.07644	-0.13929	0.41039
115	3	1.000	-2.10	-0.00	0.853	0.853	0.847	-0.07389	-0.13650	0.40007
115	4	1.000	-1.05	-0.00	0.856	0.856	0.850	-0.07090	-0.13408	0.39278
115	5	1.000	-0.57	-0.00	0.857	0.857	0.856	-0.07335	-0.13256	0.39333
115	6	1.001	0.42	-0.00	0.858	0.858	0.850	-0.07249	-0.13237	0.39742
115	7	0.996	1.05	-0.00	0.859	0.861	0.852	-0.07256	-0.13178	0.40728
115	8	0.999	2.11	-0.00	0.855	0.858	0.846	-0.07450	-0.13478	0.42150
115	9	0.997	3.12	-0.00	0.852	0.858	0.843	-0.07622	-0.13792	0.43459
115	10	0.997	4.12	-0.00	0.842	0.852	0.834	-0.08164	-0.14351	0.43951
115	11	0.999	-0.03	-0.00	0.856	0.855	0.848	-0.07399	-0.13514	0.43951
115	12	0.999	-0.03	-0.00	0.856	0.855	0.848	-0.07399	-0.13514	0.43951
117	3	0.997	4.09	0.00	3.621	2.73	1.93	0.18754	0.26045	0.32873
117	4	0.999	3.07	-0.00	3.667	2.93	1.306	0.18561	0.27264	0.33497
117	5	0.999	2.07	-0.00	3.552	3.15	1.321	0.18910	0.29070	0.34976
117	6	0.999	1.06	-0.00	3.557	3.18	1.313	0.18170	0.29913	0.33702
117	7	0.996	-0.03	-0.00	3.557	3.19	1.311	0.18124	0.28969	0.34056
117	8	0.996	-0.03	-0.00	3.557	3.19	1.311	0.18124	0.28969	0.34117
117	9	1.000	0.43	-0.00	3.615	3.24	1.303	0.18776	0.28702	0.34164
117	10	0.999	0.99	-0.00	3.615	3.24	1.303	0.19117	0.28774	0.34103
117	11	0.999	2.06	-0.00	3.615	3.24	1.303	0.19791	0.26682	0.33433
117	12	0.998	3.06	-0.00	3.687	3.07	1.243	0.20516	0.23892	0.33351

TABLE VB
BALANCE CAVITY AND MODEL BASE PRESSURE COEFFICIENTS AND INCREMENTAL
AXIAL FORCE COEFFICIENTS - THRUST EFFECT PHASE

Run	Pt.	M _c	α	Φ	C _{pC}	C _{pB1}	C _{pB2}	ΔC _{A C}	ΔC _{A B}	C _{A U}
117	13	0.999	4.06	-0.00	1.407	1.258	1.214	0.20985	0.21627	-0.32491
117	14	0.997	-0.05	-0.00	1.362	1.327	1.320	0.18727	0.29765	-0.34409
118	1	1.001	4.04	-0.00	1.492	1.327	1.384	0.25263	0.32424	-0.48734
118	2	1.001	-3.03	-0.00	1.499	1.341	1.412	0.25580	0.34361	-0.51037
118	3	0.998	-2.07	-0.00	1.489	1.380	1.431	0.25266	0.37274	-0.52034
118	4	0.998	-1.57	-0.00	1.479	1.421	1.441	0.24802	0.39587	-0.52913
118	5	1.000	-0.57	-0.00	1.480	1.450	1.452	0.24611	0.41218	-0.53185
118	6	0.999	-0.05	-0.00	1.481	1.442	1.448	0.24829	0.40814	-0.53528
118	7	0.998	0.43	-0.00	1.498	1.409	1.411	0.25665	0.37610	-0.53592
118	8	0.997	0.99	-0.00	1.503	1.368	1.368	0.26060	0.33888	-0.52939
118	9	0.996	2.04	-0.00	1.536	1.252	1.244	0.17405	0.21248	-0.27890
118	10	0.999	3.09	-0.00	1.349	1.240	1.224	0.17857	0.21543	-0.25149
118	11	0.999	4.09	-0.00	1.359	1.216	1.215	0.18493	0.19439	-0.25149
118	12	1.000	1.99	-0.00	1.532	1.325	1.325	0.27303	0.27589	-0.49063
118	13	1.000	3.04	-0.00	1.537	1.301	1.253	0.27353	0.25657	-0.49063
118	14	0.994	4.06	-0.00	1.537	1.301	1.253	0.27353	0.25657	-0.49063
118	15	0.997	-0.05	-0.00	1.472	1.436	1.439	0.24452	0.40299	-0.52636
119	4	0.999	4.09	-0.00	1.239	1.165	1.191	0.12331	0.17233	-0.15747
119	5	0.996	-3.07	-0.00	1.214	1.165	1.170	0.10960	0.15502	-0.13134
119	6	0.999	-2.08	-0.00	1.178	1.176	1.176	0.11067	0.16168	-0.13134
119	7	0.999	-1.08	-0.00	1.179	1.148	1.132	0.09198	0.12863	-0.07555
119	8	0.998	-0.55	-0.00	1.155	1.142	1.129	0.09233	0.12590	-0.07555
119	9	0.998	0.46	-0.00	1.135	1.101	1.085	0.07020	0.08598	-0.01190
119	10	0.999	0.97	-0.00	1.135	1.101	1.085	0.06994	0.08553	-0.01190
119	11	0.998	2.04	-0.00	1.036	1.016	1.007	0.01871	0.01496	-0.04359
119	12	0.997	3.07	-0.00	1.029	1.008	1.008	0.01501	0.00226	-0.04359
119	13	0.998	4.09	-0.00	1.020	1.000	1.004	0.01066	0.00226	-0.04359
119	14	0.998	-0.04	-0.00	0.912	0.895	0.891	0.04505	0.09795	-0.11149
120	1	0.999	4.08	-0.00	1.101	0.77	0.70	0.05213	0.0806	-0.04893
120	2	0.997	-3.08	-0.00	0.966	0.943	0.928	0.06355	0.08710	-0.04893
120	3	0.997	-2.08	-0.00	0.962	0.941	0.923	0.01725	0.05207	-0.02510
120	4	0.998	-1.56	-0.00	0.964	0.944	0.927	0.01925	0.05207	-0.02510
120	5	0.998	0.55	-0.00	0.922	0.943	0.930	0.01851	0.05064	-0.02510
120	6	1.000	0.47	-0.00	0.962	0.885	0.883	0.01951	0.05537	-0.02510
120	7	0.999	1.03	-0.00	0.897	0.885	0.881	0.05215	0.10623	-0.03476
120	8	0.999	3.08	-0.00	0.895	0.875	0.879	0.05403	0.11350	-0.03476
120	9	1.000	-0.08	-0.00	0.892	0.875	0.876	0.05529	0.11350	-0.03476

TABLE VB
BALANCE CAVITY AND MODEL BASE PRESSURE COEFFICIENTS AND INCREMENTAL
AXIAL FORCE COEFFICIENTS - THRUST EFFECT PHASE

Run	Pt.	M _c	α	Φ	C _{Pc}	C _{Pb1}	C _{Pb2}	ΔC _{Ac}	ΔC _{Ab}	C _{Au}
120	11	0.997	4.09	-0.00	0.888	0.872	0.870	-0.05758	-0.11841	0.37084
120	12	0.998	-0.02	-0.00	0.893	0.883	0.881	-0.05480	-0.10796	0.34465
121	1	0.997	4.08	-0.00	1.099	1.077	1.068	0.05146	0.06685	0.06057
121	2	0.999	-3.07	-0.00	1.140	1.113	1.106	0.07236	0.10078	0.01116
121	3	0.999	-2.07	-0.00	1.159	1.132	1.120	0.08228	0.11569	0.04368
121	4	0.998	-1.07	-0.00	1.159	1.132	1.112	0.08247	0.11262	0.04782
121	5	0.999	-0.55	-0.00	1.163	1.130	1.114	0.08341	0.11213	0.05005
121	6	0.996	-0.02	-0.00	1.163	1.127	1.114	0.08431	0.11066	0.04745
121	7	0.997	0.45	-0.00	1.203	1.160	1.150	0.10555	0.14322	0.07745
121	8	1.000	1.04	-0.00	1.234	1.183	1.169	0.12030	0.16705	0.13598
121	9	0.997	3.09	-0.00	1.270	1.194	1.189	0.13987	0.17614	0.18851
121	10	0.998	4.06	-0.00	1.289	1.194	1.189	0.14928	0.17614	0.18851
121	11	0.995	-0.06	-0.00	1.342	1.205	1.196	0.17774	0.25556	0.23095
121	12	0.998	-0.05	-0.00	1.327	1.286	1.273	0.16904	0.25663	0.28449
122	4	0.897	4.03	-0.00	0.926	0.925	0.923	0.04696	0.08600	0.27879
122	5	0.897	-3.06	-0.00	0.925	0.925	0.922	0.04736	0.08634	0.27819
122	6	0.902	-2.08	-0.00	0.929	0.928	0.924	0.04433	0.08217	0.27197
122	7	0.901	-1.05	-0.00	0.929	0.929	0.925	0.04441	0.08148	0.27197
122	8	0.898	-0.03	-0.00	0.929	0.928	0.924	0.04499	0.08160	0.26952
122	9	0.899	0.48	-0.00	0.927	0.927	0.923	0.04620	0.08295	0.27575
122	10	0.901	1.02	-0.00	0.930	0.930	0.925	0.04400	0.08065	0.27575
122	11	0.903	2.02	-0.00	0.928	0.930	0.924	0.04480	0.08065	0.27575
122	12	0.907	3.06	-0.00	0.928	0.934	0.924	0.04559	0.07902	0.27972
122	13	0.901	4.04	-0.00	0.928	0.936	0.922	0.04549	0.07902	0.27972
122	14	0.901	-0.04	-0.00	0.928	0.928	0.925	0.04510	0.08225	0.27177
122	15	0.901	-0.04	-0.00	0.928	0.928	0.925	0.04510	0.08225	0.27177
123	1	0.901	4.02	-0.00	1.208	1.169	1.165	0.13199	0.18389	0.21956
123	2	0.899	-3.07	-0.00	1.204	1.173	1.172	0.13015	0.19228	0.22428
123	3	0.900	-2.05	-0.00	1.203	1.176	1.172	0.12987	0.19528	0.22428
123	4	0.900	-1.05	-0.00	1.203	1.172	1.166	0.12914	0.19641	0.21889
123	5	0.902	-0.03	-0.00	1.204	1.171	1.163	0.13098	0.19782	0.21606
123	6	0.899	0.49	-0.00	1.207	1.166	1.162	0.13072	0.18539	0.21169
123	7	0.901	1.04	-0.00	1.206	1.163	1.157	0.13072	0.18044	0.21169
123	8	0.900	2.04	-0.00	1.210	1.154	1.150	0.13321	0.18044	0.20040
123	9	0.899	3.04	-0.00	1.184	1.125	1.119	0.13748	0.17201	0.14540
123	10	0.899	4.08	-0.00	1.185	1.125	1.119	0.13748	0.17201	0.14540
123	11	0.901	-0.04	-0.00	1.179	1.145	1.135	0.12353	0.15767	0.15767
123	12	0.901	-0.04	-0.00	1.179	1.145	1.135	0.12353	0.15767	0.15767

TABLE VB
BALANCE CAVITY AND MODEL BASE PRESSURE COEFFICIENTS AND INCREMENTAL
AXIAL FORCE COEFFICIENTS - THRUST EFFECT PHASE

Run	Pt.	M _c	α	φ	C _{pC}	C _{pbl}	C _{pbl2}	ΔC _{Ac}	ΔC _{Ab}	ΔC _{Au}
124	1	0.898	-4.05	-0.00	1.188	1.144	1.147	0.11976	0.16507	-0.18733
124	2	0.900	-3.04	-0.00	1.191	1.152	1.156	0.12131	0.17390	-0.19056
124	3	0.902	-2.05	-0.00	1.189	1.161	1.155	0.11922	0.17681	-0.19076
124	4	0.900	-1.05	-0.00	1.187	1.156	1.147	0.11908	0.17718	-0.18603
124	5	0.900	-0.53	-0.00	1.189	1.154	1.144	0.11908	0.17164	-0.18202
124	6	0.900	-0.02	-0.00	1.186	1.150	1.142	0.12020	0.16861	-0.18440
124	7	0.902	1.04	-0.00	1.190	1.150	1.142	0.11853	0.16535	-0.18202
124	8	0.902	2.04	-0.00	1.191	1.141	1.135	0.12064	0.16332	-0.17943
124	9	0.902	3.06	-0.00	1.195	1.132	1.128	0.12354	0.15666	-0.17272
124	10	0.901	4.09	-0.00	1.161	1.096	1.109	0.12354	0.14666	-0.16172
124	11	0.899	-0.02	-0.00	1.151	1.118	1.108	0.10964	0.10663	-0.10544
124	12	0.899	-0.02	-0.00	1.151	1.118	1.108	0.09640	0.12840	-0.11902
125	1	0.899	-4.03	-0.00	1.350	1.245	1.278	0.22295	0.29608	-0.44166
125	2	0.898	-3.04	-0.00	1.352	1.269	1.296	0.22336	0.31487	-0.45821
125	3	0.899	-2.08	-0.00	1.349	1.269	1.303	0.22219	0.33484	-0.46683
125	4	0.897	-1.03	-0.00	1.348	1.315	1.324	0.22196	0.36294	-0.48055
125	5	0.902	-0.54	-0.00	1.349	1.328	1.333	0.22082	0.37190	-0.48055
125	6	0.900	-0.01	-0.00	1.351	1.323	1.327	0.22314	0.36875	-0.48091
125	7	0.898	0.59	-0.00	1.346	1.323	1.327	0.22198	0.36801	-0.48061
125	8	0.897	1.59	-0.00	1.374	1.287	1.277	0.23380	0.37066	-0.46458
125	9	0.901	2.57	-0.00	1.365	1.241	1.235	0.23525	0.37066	-0.44575
125	10	0.900	3.57	-0.00	1.396	1.224	1.191	0.24514	0.24067	-0.44729
125	11	0.900	4.57	-0.00	1.350	1.224	1.191	0.25143	0.22312	-0.44729
125	12	0.900	-0.03	-0.00	1.350	1.326	1.326	0.22266	0.36810	-0.48748
126	3	0.404	-4.00	-0.00	0.987	0.987	0.987	0.3883	0.7113	0.25857
126	4	0.400	-3.01	-0.00	0.987	0.986	0.987	0.3903	0.7138	0.26486
126	5	0.401	-2.08	-0.00	0.987	0.987	0.987	0.3848	0.7133	0.27472
126	6	0.400	-1.03	-0.00	0.987	0.987	0.987	0.4040	0.7215	0.27808
126	7	0.401	-0.56	-0.00	0.988	0.987	0.987	0.3784	0.6889	0.27218
126	8	0.402	0.45	-0.00	0.987	0.987	0.987	0.3865	0.7025	0.28193
126	9	0.401	1.45	-0.00	0.987	0.987	0.987	0.3952	0.6821	0.27719
126	10	0.402	2.42	-0.00	0.988	0.988	0.987	0.3932	0.6721	0.27719
126	11	0.402	3.40	-0.00	0.988	0.988	0.987	0.3767	0.6925	0.27165
126	12	0.401	4.35	-0.00	0.987	0.987	0.987	0.3933	0.6925	0.27165
126	13	0.401	-0.03	-0.00	0.987	0.987	0.987	0.3933	0.6925	0.27165
126	14	0.401	-0.04	-0.00	0.988	0.987	0.987	0.3768	0.6995	0.26943
127	6	0.005	0.00	0.00	0.793	0.793	0.793	0.07138	0.1313	0.50878
127	7	0.999	-0.00	-0.00	0.861	0.859	0.853	0.07138	0.1313	0.39136
127	8	1.001	-0.01	-0.00	1.140	1.106	1.090	0.07177	0.08985	0.00990

TABLE VB
BALANCE CAVITY AND MODEL BASE PRESSURE COEFFICIENTS AND INCREMENTAL AXIAL FORCE COEFFICIENTS - THRUST EFFECT PHASE

Run	Pt.	M _c	α	Φ	C _p c	C _p b ₁	C _p b ₂	ΔC _A c	ΔC _A b	C _A u
127	9	0.999	-0.03	-0.00	1.362	1.218	1.201	0.13528	0.19229	-0.16718
127	10	0.997	-0.04	-0.00	1.319	1.276	1.264	0.16519	0.24866	-0.27790
127	11	1.000	-0.02	-0.00	1.385	1.385	1.383	0.19701	0.31547	-0.37528
127	13	0.998	-0.03	-0.00	1.453	1.420	1.445	0.21590	0.35177	-0.43445
127	14	0.992	-0.02	-0.00	1.480	1.439	1.445	0.25081	0.38530	-0.49470
127	15	0.956	-0.05	-0.00	0.961	0.964	0.951	0.02732	0.04296	-0.42431
127	17	0.997	-0.03	-0.00	1.521	1.468	0.972	0.01733	0.03493	-0.43413
127	18	0.997	-0.02	-0.00	1.530	1.475	1.480	0.26877	0.43469	-0.59289
127	19	0.952	-0.03	-0.00	1.473	1.475	1.440	0.27414	0.44338	-0.60754
127	19	0.952	-0.03	-0.00	1.473	1.475	1.440	0.26853	0.43740	-0.60256
128	6	1.103	4.05	-0.00	1.532	1.297	1.391	0.25511	0.58663	-0.34746
128	7	1.100	-4.05	-0.00	1.518	1.307	1.388	0.22036	0.26166	-0.33579
128	8	1.101	-2.08	-0.00	1.526	1.317	1.412	0.22273	0.27554	-0.34732
128	9	1.096	-1.05	-0.00	1.512	1.358	1.451	0.21913	0.30563	-0.35830
128	10	1.099	-0.56	-0.00	1.525	1.423	1.464	0.21905	0.35077	-0.39222
128	11	1.101	0.25	-0.00	1.522	1.462	1.476	0.22181	0.35401	-0.38964
128	12	1.102	0.49	-0.00	1.561	1.407	1.428	0.23027	0.31487	-0.38664
128	14	1.096	2.07	-0.00	1.568	1.365	1.372	0.23301	0.28413	-0.36750
128	15	1.100	3.07	-0.00	1.580	1.329	1.355	0.24658	0.21052	-0.34787
128	16	1.100	4.07	-0.00	1.598	1.317	1.221	0.24588	0.20559	-0.35069
128	17	1.097	3.07	-0.00	1.520	1.307	1.221	0.22155	0.20004	-0.35069
128	18	1.098	-0.04	-0.00	1.520	1.456	1.468	0.22155	0.35000	-0.35069
129	3	0.396	-4.01	-0.00	1.068	1.050	1.053	0.22441	0.30256	-0.41771
129	4	0.396	-3.06	-0.00	1.070	1.054	1.056	0.22664	0.32230	-0.43391
129	5	0.397	-1.02	-0.00	1.069	1.057	1.053	0.22948	0.33422	-0.44558
129	6	0.397	-0.55	-0.00	1.068	1.064	1.065	0.22390	0.36796	-0.46218
129	7	0.397	-0.45	-0.00	1.071	1.063	1.064	0.22706	0.37944	-0.47994
129	8	0.396	0.98	-0.00	1.073	1.059	1.060	0.23420	0.34772	-0.45296
129	10	0.398	1.95	-0.00	1.074	1.049	1.056	0.24451	0.31701	-0.43725
129	11	0.394	3.00	-0.00	1.074	1.047	1.051	0.24455	0.28400	-0.41169
129	11	0.395	4.00	-0.00	1.075	1.041	1.050	0.24516	0.26586	-0.41169
130	1	0.797	4.02	-0.00	1.288	1.180	1.219	0.33371	0.28715	-0.44926
130	2	0.801	-3.07	-0.00	1.291	1.192	1.236	0.23544	0.30610	-0.47004
130	3	0.797	-1.04	-0.00	1.280	1.221	1.252	0.23371	0.33731	-0.47554
130	4	0.797	-1.04	-0.00	1.280	1.247	1.254	0.22641	0.36069	-0.47960

TABLE VB
BALANCE CAVITY AND MODEL BASE PRESSURE COEFFICIENTS AND INCREMENTAL
AXIAL FORCE COEFFICIENTS - THRUST EFFECT PHASE

Run	Pt.	M _c	α	φ	C _{Pc}	C _{Pb1}	C _{Pb2}	ΔC _{Ac}	ΔC _{Ab}	C _{Au}
130	5	0.800	-0.55	-0.00	1.279	1.261	1.263	0.22429	0.37420	-0.48463
130	6	0.801	-0.046	-0.00	1.284	1.260	1.260	0.22709	0.36982	-0.48279
130	8	0.799	0.98	-0.00	1.292	1.205	1.205	0.23474	0.35482	-0.48376
130	10	0.797	1.98	-0.00	1.310	1.183	1.183	0.24198	0.29506	-0.47184
130	11	0.795	3.02	-0.00	1.314	1.177	1.151	0.25577	0.26113	-0.45354
130		0.798	4.02	-0.00	1.325	1.171	1.142	0.26235	0.22551	-0.44786
131	1	0.799	4.06	-0.00	1.164	1.127	1.129	0.13199	0.18318	-0.21481
131	2	0.799	-3.08	-0.00	1.167	1.133	1.136	0.13477	0.19349	-0.21932
131	3	0.795	-2.04	-0.00	1.164	1.140	1.134	0.13135	0.19667	-0.21853
131	4	0.799	-0.55	-0.00	1.165	1.139	1.135	0.13227	0.19902	-0.21959
131	5	0.797	-0.05	-0.00	1.166	1.137	1.131	0.13374	0.19707	-0.21963
131	6	0.799	0.45	-0.00	1.169	1.130	1.134	0.13459	0.19276	-0.21872
131	8	0.791	0.99	-0.00	1.172	1.137	1.131	0.13590	0.19482	-0.21904
131	9	0.796	1.99	-0.00	1.179	1.129	1.123	0.13762	0.19256	-0.21636
131	10	0.801	3.02	-0.00	1.179	1.124	1.120	0.14415	0.18256	-0.21220
131	11	0.798	4.02	-0.00	1.182	1.111	1.112	0.14740	0.17472	-0.21220
132	3	1.245	4.07	-0.00	1.571	1.279	1.279	0.18942	0.19802	-0.23334
132	4	1.251	-3.08	-0.00	1.606	1.254	1.254	0.19814	0.17758	-0.25641
132	5	1.248	-2.06	-0.00	1.577	1.235	1.235	0.19525	0.23506	-0.26819
132	6	1.254	-0.56	-0.00	1.577	1.235	1.235	0.19068	0.26597	-0.28310
132	8	1.253	0.43	-0.00	1.588	1.248	1.248	0.19877	0.27897	-0.28323
132	9	1.250	0.98	-0.00	1.610	1.248	1.248	0.20467	0.25893	-0.28231
132	10	1.251	0.25	-0.00	1.624	1.258	1.258	0.20463	0.24424	-0.28457
132	11	1.252	3.02	-0.00	1.682	1.220	1.220	0.21593	0.21244	-0.26536
132		1.248	4.02	-0.00	1.695	1.293	1.267	0.22968	0.19755	-0.26536
133	1	1.257	4.06	-0.00	1.388	1.257	1.284	0.12733	0.15744	-0.09379
133	2	1.247	-3.07	-0.00	1.377	1.262	1.280	0.12473	0.16667	-0.09416
133	3	1.250	-2.07	-0.00	1.383	1.230	1.230	0.12515	0.17455	-0.09374
133	4	1.248	-0.56	-0.00	1.371	1.210	1.210	0.12281	0.18614	-0.09330
133	5	1.248	0.45	-0.00	1.368	1.213	1.213	0.12172	0.18533	-0.09210
133	6	1.248	0.05	-0.00	1.366	1.214	1.214	0.12470	0.18533	-0.09210
133	7	1.250	0.01	-0.00	1.366	1.214	1.214	0.12750	0.18029	-0.09269
133	8	1.251	1.06	-0.00	1.417	1.261	1.285	0.13273	0.16725	-0.09089
133	9	1.251	2.07	-0.00	1.417	1.261	1.285	0.13273	0.15975	-0.09089
133	10	1.251	3.07	-0.00	1.417	1.261	1.285	0.13273	0.15975	-0.09089

TABLE VB
BALANCE CAVITY AND MODEL BASE PRESSURE COEFFICIENTS AND INCREMENTAL AXIAL FORCE COEFFICIENTS - THRUST EFFECT PHASE

Run	Pt.	M _c	α	Φ	C _{p c}	C _{p b1}	C _{p b2}	ΔC _{A c}	ΔC _{A b}	C _{A u}
133	11	1.249	4.12	-0.00	1.427	1.239	1.284	0.14099	0.15362	-0.08601
134	1	1.250	-4.10	-0.00	1.290	1.208	1.217	0.09562	0.12447	-0.001234
134	2	1.250	-2.13	-0.00	1.284	1.221	1.222	0.09354	0.13012	-0.00356
134	3	1.250	-1.07	-0.00	1.268	1.222	1.222	0.08850	0.13326	0.00049
134	4	1.251	-0.56	-0.00	1.272	1.228	1.225	0.08759	0.13126	0.00111
134	5	1.251	-0.05	-0.00	1.268	1.226	1.225	0.08927	0.13372	0.00178
134	6	1.251	0.03	-0.00	1.272	1.226	1.231	0.08835	0.13178	0.00324
134	7	1.252	0.41	-0.00	1.281	1.217	1.224	0.09269	0.13363	0.00234
134	8	1.252	1.03	-0.00	1.291	1.199	1.212	0.09561	0.12018	0.00128
134	9	1.249	3.10	-0.00	1.296	1.174	1.199	0.09561	0.12018	0.00575
134	10	1.250	4.12	-0.00	1.306	1.162	1.202	0.10067	0.10973	0.00605
134	11	1.250								
136	2	1.249	4.16	-0.00	1.570	1.316	1.317	0.18795	0.18579	-0.18650
136	3	1.251	-2.16	-0.00	1.567	1.411	1.403	0.18643	0.23744	-0.21803
136	4	1.249	-1.57	-0.00	1.570	1.464	1.439	0.18812	0.26397	-0.21991
136	5	1.251	-0.55	-0.00	1.562	1.455	1.468	0.18816	0.27903	-0.24148
136	6	1.251	0.46	-0.00	1.599	1.416	1.516	0.18839	0.28376	-0.24355
136	7	1.250	1.06	-0.00	1.613	1.394	1.501	0.19674	0.26813	-0.24855
136	8	1.250	3.15	-0.00	1.607	1.357	1.478	0.20929	0.25527	-0.24892
136	9	1.249	4.19	-0.00	1.625	1.304	1.369	0.19457	0.25739	-0.22705
136	10	1.249								
136	11	1.249								
137	1	1.249	4.15	-0.00	1.410	1.298	1.275	0.13499	0.16792	-0.07884
137	2	1.251	-2.19	-0.00	1.406	1.320	1.350	0.13342	0.19596	-0.10497
137	3	1.251	-1.09	-0.00	1.420	1.352	1.354	0.13487	0.19639	-0.10452
137	4	1.251	-0.57	-0.00	1.430	1.368	1.355	0.13866	0.21098	-0.12037
137	5	1.251	0.47	-0.00	1.429	1.351	1.356	0.14126	0.21136	-0.12102
137	6	1.250	1.07	-0.00	1.434	1.339	1.350	0.14280	0.20171	-0.11846
137	7	1.250	3.12	-0.00	1.442	1.321	1.345	0.14536	0.20630	-0.11568
137	8	1.251	4.18	-0.00	1.449	1.305	1.345	0.14614	0.19981	-0.10951
137	9	1.251								
137	10	1.251								
137	11	1.251								
138	1	1.252	4.18	-0.00	1.321	1.270	1.240	0.14771	0.17289	-0.08575
138	2	1.254	-2.15	-0.00	1.320	1.263	1.240	0.14998	0.14674	-0.02888
138	3	1.250	-1.07	-0.00	1.320	1.269	1.271	0.10506	0.15712	-0.02425
138	4	1.249	-0.57	-0.00	1.322	1.277	1.266	0.10535	0.16051	-0.03082
138	5	1.254	-0.04	-0.00	1.322	1.277	1.263	0.10595	0.15944	-0.03406
138	6	1.254	0.47	-0.00	1.325	1.277	1.263	0.10643	0.15715	-0.02891
138	7	1.254								
138	8	1.254								
138	9	1.254								
138	10	1.254								
138	11	1.254								

TABLE VB
BALANCE CAVITY AND MODEL BASE PRESSURE COEFFICIENTS AND INCREMENTAL
AXIAL FORCE COEFFICIENTS - THRUST EFFECT PHASE

Run	Pt.	M _c	α	Φ	C _{p c}	C _{p b1}	C _{p b2}	ΔC _{A c}	ΔC _{A b}	C _{A u}
138	7	1.252	1.01	-0.00	1.325	1.269	1.260	0.10665	0.15440	-0.02766
138	8	1.252	2.07	-0.00	1.323	1.253	1.252	0.10621	0.14733	-0.01938
138	9	1.251	3.15	-0.00	1.317	1.237	1.244	0.10426	0.14072	-0.01352
138	10	1.249	4.20	-0.00	1.308	1.227	1.232	0.10172	0.13465	-0.00091
139	1	1.251	-4.18	-0.00	0.776	0.785	0.766	-0.07336	-0.13066	0.45833
139	2	1.249	-2.13	-0.00	0.778	0.783	0.772	-0.07290	-0.12997	0.44980
139	3	1.250	-1.05	-0.00	0.781	0.784	0.776	-0.07182	-0.12816	0.44713
139	4	1.251	-0.56	-0.00	0.781	0.784	0.777	-0.07165	-0.12799	0.44495
139	5	1.252	-0.04	-0.00	0.780	0.782	0.777	-0.07146	-0.12752	0.44451
139	6	1.251	0.47	-0.00	0.780	0.782	0.776	-0.07205	-0.12871	0.44589
139	7	1.250	1.01	-0.00	0.777	0.779	0.775	-0.07226	-0.12972	0.44310
139	8	1.250	3.15	-0.00	0.776	0.779	0.773	-0.07300	-0.13073	0.45032
139	9	1.250	4.19	-0.00	0.776	0.777	0.769	-0.07340	-0.13163	0.45391
139	10	0.035	0.00	-0.00	0.785	0.786	0.762	-0.07601	-0.13460	0.46140
139	11									0.70175
140	3	1.102	-4.11	0.00	1.510	1.268	1.381	0.21601	0.25201	-0.30116
140	4	1.103	-2.12	-0.00	1.493	1.273	1.413	0.20866	0.25546	-0.30199
140	5	1.100	-1.06	-0.00	1.496	1.416	1.429	0.21057	0.25919	-0.31639
140	6	1.098	-0.59	-0.00	1.487	1.432	1.458	0.20778	0.25965	-0.33539
140	7	1.100	-0.05	-0.00	1.486	1.434	1.455	0.20717	0.25965	-0.35105
140	8	1.099	0.44	-0.00	1.508	1.404	1.433	0.21597	0.25947	-0.34775
140	9	1.098	0.99	-0.00	1.520	1.384	1.407	0.22129	0.25947	-0.34595
140	10	1.098	2.08	-0.00	1.528	1.364	1.366	0.22308	0.27068	-0.35087
140	11	1.102	3.15	-0.00	1.570	1.323	1.334	0.24106	0.27068	-0.35167
140	12		4.15	-0.00		1.306	1.249	0.24106	0.27068	-0.35067
141	1	1.100	-4.16	-0.00	0.854	0.864	0.849	-0.06196	-0.10807	0.41821
141	2	1.101	-2.14	-0.00	0.852	0.858	0.847	-0.06255	-0.11073	0.41243
141	3	1.101	-1.08	-0.00	0.863	0.866	0.858	-0.05800	-0.11034	0.40060
141	4	1.097	-0.59	-0.00	0.864	0.866	0.860	-0.05775	-0.10337	0.40060
141	5	1.100	-0.05	-0.00	0.863	0.865	0.857	-0.05806	-0.10513	0.40629
141	6	1.102	0.45	-0.00	0.855	0.858	0.853	-0.05806	-0.10798	0.40112
141	7	1.103	1.02	-0.00	0.852	0.855	0.850	-0.06222	-0.11016	0.41612
141	8	1.102	2.09	-0.00	0.852	0.855	0.850	-0.06248	-0.11090	0.41207
141	9	1.102	3.15	-0.00	0.849	0.852	0.848	-0.06430	-0.11132	0.41199
141	10		4.18	-0.00						0.41199
142	4	1.000	-4.13	-0.00	0.887	0.895	0.882	-0.05772	-0.10149	0.35570
142	5	1.098	-2.13	-0.00	0.887	0.891	0.882	-0.05792	-0.10359	0.34868
142	6	1.001	-1.08	-0.00	0.887	0.890	0.882	-0.05755	-0.10344	0.35543

TABLE VB
BALANCE CAVITY AND MODEL BASE PRESSURE COEFFICIENTS AND INCREMENTAL
AXIAL FORCE COEFFICIENTS - THRUST EFFECT PHASE

Run	Pt.	M _c	α	Φ	C _{Pc}	C _{Pb1}	C _{Pb2}	ΔC _{Ac}	ΔC _{Ab}	C _{Au}
142	7	0.999	-0.57	-0.00	0.882	0.884	0.877	-0.06025	-0.10887	0.34987
142	8	1.000	-0.03	-0.00	0.885	0.887	0.883	-0.05871	-0.10581	0.34914
142	9	1.000	0.46	-0.00	0.889	0.890	0.883	-0.05654	-0.10242	0.35876
142	10	1.000	0.99	-0.00	0.882	0.883	0.877	-0.06044	-0.10907	0.35299
142	11	1.000	2.06	-0.00	0.889	0.891	0.885	-0.05664	-0.10118	0.36051
142	12	1.000	3.14	-0.00	0.878	0.882	0.874	-0.06230	-0.11100	0.36011
142	13	1.002	4.17	-0.00	0.880	0.884	0.877	-0.06116	-0.10850	0.37026
143	1	1.000	-4.10	0.00	1.444	1.328	1.369	0.22831	0.31872	0.43142
143	2	1.000	-2.11	-0.00	1.454	1.382	1.408	0.23363	0.36144	0.46403
143	3	0.997	-1.05	-0.00	1.455	1.406	1.414	0.23529	0.37702	0.48023
143	4	0.999	-0.57	-0.00	1.453	1.423	1.426	0.23364	0.38936	0.48128
143	5	1.000	-0.03	-0.00	1.456	1.420	1.424	0.24111	0.38759	0.48089
143	6	1.000	0.46	-0.00	1.468	1.405	1.405	0.24459	0.37032	0.48199
143	7	0.998	0.99	-0.00	1.473	1.375	1.375	0.24452	0.34467	0.48137
143	8	0.999	2.03	-0.00	1.477	1.340	1.345	0.24572	0.31375	0.46032
143	9	1.000	3.11	-0.00	1.472	1.314	1.335	0.24262	0.29645	0.44020
143	10	1.004	4.14	-0.00	1.483	1.304	1.330	0.24464	0.27449	0.42784
144	1	1.002	-4.14	0.00	1.361	1.272	1.302	0.18474	0.26152	0.30744
144	2	0.999	-2.14	-0.00	1.353	1.302	1.315	0.18220	0.28244	0.31986
144	3	1.000	-1.06	-0.00	1.356	1.322	1.316	0.18493	0.29144	0.32504
144	4	1.000	-0.58	-0.00	1.363	1.326	1.316	0.18613	0.29439	0.33042
144	5	0.999	-0.02	-0.00	1.368	1.329	1.314	0.18773	0.29378	0.32694
144	6	0.999	0.46	-0.00	1.368	1.322	1.314	0.18951	0.29189	0.32231
144	7	0.998	1.02	-0.00	1.375	1.317	1.310	0.19369	0.28744	0.31713
144	8	1.000	2.06	-0.00	1.386	1.300	1.298	0.19878	0.27441	0.30676
144	9	1.000	3.12	-0.00	1.386	1.278	1.284	0.19878	0.25707	0.30676
144	10	1.000	4.16	-0.00	1.386	1.264	1.273	0.19896	0.24568	0.29726
145	1	0.999	-4.13	0.00	1.263	1.207	1.222	0.13551	0.19360	0.18451
145	2	1.001	-2.13	-0.00	1.269	1.235	1.228	0.13799	0.20868	0.19482
145	3	1.000	-1.06	-0.00	1.272	1.239	1.218	0.14151	0.20908	0.20103
145	4	0.997	-0.57	-0.00	1.274	1.235	1.213	0.14075	0.20596	0.20438
145	5	1.003	-0.04	-0.00	1.285	1.246	1.223	0.14562	0.20542	0.20339
145	6	0.994	0.47	-0.00	1.277	1.232	1.215	0.14340	0.20408	0.19984
145	7	0.997	1.04	-0.00	1.288	1.228	1.215	0.14786	0.20332	0.19265
145	8	1.000	2.06	-0.00	1.288	1.206	1.213	0.14779	0.20328	0.19265
145	9	0.997	3.13	-0.00	1.292	1.200	1.210	0.14779	0.19328	0.18734
145	10	1.001	4.18	-0.00	1.292	1.200	1.210	0.14980	0.18693	0.17391
146	1	0.899	-4.12	-0.00	0.929	0.934	0.925	-0.04469	-0.07893	0.26614

TABLE VB
BALANCE CAVITY AND MODEL BASE PRESSURE COEFFICIENTS AND INCREMENTAL AXIAL FORCE COEFFICIENTS - THRUST EFFECT PHASE

Run	Pt.	M _c	α	Φ	C _{Pc}	C _{Pb1}	C _{Pb2}	ΔC _{Ac}	ΔC _{Ab}	C _{Au}
146	3	0.901	-2.11	-0.00	0.928	0.932	0.926	0.04511	-0.07940	0.26552
146	3	0.901	-1.07	-0.00	0.930	0.933	0.927	0.04425	-0.07798	0.26538
146	4	0.900	-0.04	-0.00	0.927	0.930	0.925	0.04621	-0.08115	0.26432
146	6	0.898	0.46	-0.00	0.927	0.930	0.926	0.04606	-0.08140	0.26659
146	7	0.902	1.01	-0.00	0.931	0.933	0.929	0.04355	-0.07677	0.26589
146	8	0.901	2.05	-0.00	0.932	0.934	0.930	0.04257	-0.07569	0.26970
146	9	0.901	3.13	-0.00	0.927	0.932	0.926	0.04570	-0.07955	0.27374
146	10	0.901	4.12	-0.00	0.926	0.930	0.924	0.04662	-0.08155	0.27704
147	3	0.898	4.09	0.00	3.553	2.50	2.95	0.22504	0.30907	0.4905
147	4	0.896	-2.06	-0.00	3.557	2.95	3.21	0.22892	0.35153	0.47694
147	5	0.898	-1.06	-0.00	3.561	3.21	3.29	0.23176	0.36865	0.48921
147	6	0.899	-0.56	-0.00	3.564	3.32	3.39	0.23399	0.38244	0.49490
147	7	0.898	0.01	-0.00	3.572	3.32	3.38	0.23175	0.37918	0.49351
147	8	0.898	0.45	-0.00	3.374	3.10	3.18	0.23698	0.35652	0.49090
147	9	0.897	1.06	-0.00	3.374	2.87	2.92	0.23081	0.32901	0.47127
147	10	0.897	2.06	-0.00	3.371	2.65	2.68	0.23370	0.30118	0.45005
147	11	0.897	3.08	-0.00	3.381	2.45	2.57	0.23400	0.29157	0.45664
147	12	0.896	4.13	-0.00	3.381	2.45	2.57	0.23400	0.28601	0.45660
148	1	0.899	-4.10	-0.00	4.219	1.72	1.75	0.13612	0.19674	0.2150
148	2	0.899	-2.10	-0.00	4.223	1.83	1.75	0.13929	0.20370	0.22965
148	3	0.899	-1.05	-0.00	4.226	1.93	1.78	0.14199	0.20947	0.23523
148	4	0.899	-0.56	-0.00	4.227	1.91	1.76	0.14213	0.20751	0.23380
148	5	0.898	0.02	-0.00	4.227	1.89	1.76	0.14474	0.20606	0.23490
148	6	0.898	0.48	-0.00	4.231	1.89	1.75	0.14457	0.20690	0.23468
148	7	0.901	1.05	-0.00	4.232	1.89	1.80	0.14690	0.20810	0.23673
148	8	0.901	2.05	-0.00	4.235	1.80	1.78	0.14690	0.20221	0.22673
148	9	0.901	3.10	-0.00	4.235	1.72	1.77	0.14869	0.19709	0.22140
148	10	0.902	4.14	-0.00	4.235	1.61	1.71	0.14897	0.18709	0.21406
149	1	0.900	-4.12	-0.00	4.185	1.50	1.47	0.17221	0.16775	0.17268
149	2	0.900	-2.10	-0.00	4.195	1.62	1.45	0.17221	0.17296	0.18222
149	3	0.903	-1.07	-0.00	4.198	1.62	1.45	0.17408	0.17387	0.18222
149	4	0.901	-0.56	-0.00	4.198	1.64	1.50	0.17564	0.17680	0.18456
149	5	0.899	0.02	-0.00	4.196	1.60	1.44	0.17564	0.17571	0.18278
149	6	0.901	1.05	-0.00	4.200	1.62	1.47	0.17564	0.17249	0.18420
149	7	0.899	2.05	-0.00	4.197	1.56	1.43	0.17564	0.17000	0.18023
149	8	0.900	3.05	-0.00	4.198	1.56	1.44	0.17564	0.17000	0.17558

TABLE VB
BALANCE CAVITY AND MODEL BASE PRESSURE COEFFICIENTS AND INCREMENTAL AXIAL FORCE COEFFICIENTS - THRUST EFFECT PHASE

Run	Pt.	M _c	α	φ	C _p c	C _p b ₁	C _p b ₂	ΔC _A c	ΔC _A b	C _A u
149	10	0.901	3.12	-0.00	1.201	1.153	1.149	0.12777	0.17074	-0.17417
149	11	0.899	4.16	-0.00	1.197	1.139	1.143	0.12553	0.15966	-0.16955
150	1	0.799	-4.06	0.00	1.273	1.192	1.225	0.22000	0.29895	-0.42395
150	2	0.800	-2.08	-0.00	1.281	1.227	1.247	0.22569	0.33878	-0.44474
150	3	0.799	-1.05	-0.00	1.281	1.247	1.250	0.22693	0.35646	-0.45719
150	4	0.799	-0.56	-0.00	1.281	1.260	1.259	0.22478	0.37202	-0.46265
150	5	0.800	-0.43	-0.00	1.281	1.254	1.259	0.22568	0.36691	-0.46163
150	6	0.800	0.46	-0.00	1.289	1.236	1.240	0.23283	0.34039	-0.45515
150	7	0.800	1.03	-0.00	1.295	1.212	1.222	0.23742	0.31016	-0.44599
150	8	0.799	1.99	-0.00	1.292	1.201	1.206	0.23522	0.29164	-0.42564
150	9	0.799	3.09	-0.00	1.290	1.187	1.200	0.23363	0.28134	-0.42317
150	10	0.799	4.10	-0.00	1.298	1.167	1.200	0.24032	0.27702	-0.43175
151	1	0.802	-4.09	0.00	1.174	1.356	1.382	0.13666	0.19370	-0.21596
151	2	0.801	-2.12	-0.00	1.179	1.456	1.422	0.13996	0.20976	-0.22927
151	3	0.802	-1.06	-0.00	1.179	1.523	1.423	0.14348	0.20969	-0.22977
151	4	0.802	-0.55	-0.00	1.181	1.531	1.443	0.14511	0.21072	-0.23099
151	5	0.800	-0.01	-0.00	1.182	1.549	1.442	0.14422	0.20728	-0.22952
151	6	0.802	0.46	-0.00	1.183	1.549	1.442	0.14569	0.20711	-0.22953
151	7	0.802	1.04	-0.00	1.184	1.548	1.441	0.14649	0.20628	-0.22994
151	8	0.802	1.85	-0.00	1.185	1.545	1.441	0.14719	0.20364	-0.22722
151	9	0.802	3.08	-0.00	1.185	1.538	1.440	0.14807	0.19796	-0.21972
151	10	0.799	4.12	-0.00	1.181	1.522	1.440	0.14803	0.17896	-0.20972
152	1	0.803	-4.08	0.00	1.129	1.105	1.093	0.10298	0.14008	-0.15503
152	2	0.803	-2.06	-0.00	1.135	1.104	1.091	0.10699	0.13999	-0.15506
152	3	0.802	-1.05	-0.00	1.135	1.105	1.093	0.10797	0.14044	-0.15521
152	4	0.802	-0.54	-0.00	1.135	1.103	1.094	0.10825	0.14192	-0.15519
152	5	0.801	-0.07	-0.00	1.134	1.101	1.091	0.10809	0.13893	-0.15520
152	6	0.801	0.47	-0.00	1.131	1.100	1.092	0.10546	0.13602	-0.15523
152	7	0.801	1.02	-0.00	1.131	1.101	1.088	0.10546	0.13479	-0.15523
152	8	0.805	1.33	-0.00	1.131	1.101	1.093	0.10575	0.13381	-0.15523
152	9	0.805	3.12	-0.00	1.131	1.103	1.092	0.10463	0.13359	-0.15523
152	10	0.803	4.12	-0.00	1.131	1.109	1.092	0.10463	0.13359	-0.15523
153	1	0.801	-0.03	0.00	0.946	0.947	0.943	-0.04243	-0.07768	0.26755
153	2	0.803	-4.11	-0.00	0.947	0.950	0.943	-0.04219	-0.07561	0.26755
153	3	0.800	-2.05	-0.00	0.944	0.945	0.941	-0.04221	-0.08043	0.26647
153	4	0.801	-1.05	-0.00	0.946	0.947	0.943	-0.04286	-0.07741	0.26647
153	5	0.801	-0.56	-0.00	0.948	0.949	0.945	-0.04084	-0.07442	0.26647
153	6	0.803	-0.07	-0.00	0.949	0.949	0.945	-0.04115	-0.07580	0.26667
153	7	0.803	0.47	-0.00	0.949	0.947	0.945	-0.04115	-0.07580	0.26667
153	8	0.803	1.02	-0.00	0.949	0.947	0.945	-0.04115	-0.07580	0.26667
153	9	0.803	1.33	-0.00	0.949	0.947	0.945	-0.04115	-0.07580	0.26667
153	10	0.803	3.12	-0.00	0.949	0.947	0.945	-0.04115	-0.07580	0.26667
153	10	0.803	4.12	-0.00	0.949	0.947	0.945	-0.04115	-0.07580	0.26667

TABLE VB
BALANCE CAVITY AND MODEL BASE PRESSURE COEFFICIENTS AND INCREMENTAL
AXIAL FORCE COEFFICIENTS - THRUST EFFECT PHASE

Run	Pt.	M _c	α	φ	C _{Pc}	C _{Pb1}	C _{Pb2}	ΔC _{Ac}	ΔC _{Ab}	C _{Au}
153	7	0.802	0.47	-0.00	0.948	0.948	0.945	-0.04144	-0.07509	0.26640
153	8	0.800	1.01	-0.00	0.945	0.945	0.942	-0.04401	-0.07968	0.26563
153	9	0.800	2.08	-0.00	0.945	0.945	0.942	-0.04415	-0.07950	0.26683
153	10	0.802	3.14	-0.00	0.947	0.949	0.945	-0.04187	-0.07483	0.26685
153	11	0.802	4.12	-0.00	0.945	0.947	0.943	-0.04355	-0.07702	0.27007
154	1	0.401	-4.04	0.00	0.988	0.988	0.988	-0.03589	-0.06611	0.26470
154	2	0.400	-2.05	-0.00	0.988	0.987	0.987	-0.03588	-0.06941	0.27406
154	3	0.401	-1.05	-0.00	0.989	0.989	0.988	-0.03464	-0.06325	0.26010
154	4	0.402	-0.55	-0.00	0.988	0.988	0.988	-0.03384	-0.06381	0.27000
154	5	0.402	-0.44	-0.00	0.988	0.988	0.988	-0.03350	-0.06451	0.26125
154	6	0.401	0.47	-0.00	0.989	0.988	0.988	-0.03384	-0.06559	0.27047
154	7	0.401	2.01	-0.00	0.989	0.988	0.988	-0.03393	-0.06572	0.27866
154	8	0.401	3.01	-0.00	0.989	0.988	0.988	-0.03375	-0.06572	0.27442
154	9	0.401	4.05	-0.00	0.988	0.988	0.988	-0.03605	-0.06723	0.27328
154	10	0.401	-4.05	-0.00	0.988	0.987	0.987	-0.03835	-0.07131	0.27246
155	1	0.399	-4.01	-0.00	1.067	1.053	1.057	0.22694	0.31791	0.37935
155	2	0.402	-2.04	-0.00	1.071	1.057	1.058	0.22695	0.32933	0.40132
155	3	0.403	-1.04	-0.00	1.069	1.063	1.063	0.22159	0.36176	0.40626
155	4	0.402	-0.55	-0.00	1.069	1.064	1.066	0.22062	0.36752	0.41580
155	5	0.402	0.45	-0.00	1.069	1.065	1.066	0.22149	0.37302	0.41700
155	6	0.402	0.97	-0.00	1.071	1.062	1.063	0.22313	0.35575	0.41020
155	7	0.397	2.01	-0.00	1.072	1.055	1.058	0.22936	0.34303	0.41087
155	8	0.400	3.04	-0.00	1.070	1.052	1.058	0.22965	0.32000	0.38616
155	9	0.400	4.04	-0.00	1.071	1.052	1.052	0.22965	0.30736	0.38186
155	10	0.402	-4.00	-0.00	1.083	1.051	1.052	0.22971	0.29609	0.35761
155	11	0.402	-4.00	-0.00	1.083	1.051	1.052	0.22971	0.29609	0.35761
157	3	1.251	4.09	0.00	1.553	1.250	1.398	0.18514	0.18947	0.20244
157	4	1.251	-2.10	-0.00	1.552	1.367	1.399	0.18117	0.18947	0.20244
157	5	1.251	-1.04	-0.00	1.544	1.401	1.429	0.17854	0.22333	0.21685
157	6	1.251	0.47	-0.00	1.544	1.427	1.438	0.17847	0.22333	0.21685
157	7	1.251	0.98	-0.00	1.568	1.375	1.431	0.18640	0.22333	0.21685
157	8	1.251	2.04	-0.00	1.588	1.346	1.390	0.19200	0.22333	0.21685
157	9	1.251	3.08	-0.00	1.598	1.322	1.344	0.19640	0.22333	0.21685
157	10	1.251	4.10	-0.00	1.622	1.298	1.314	0.20445	0.22333	0.21685
157	11	1.251	-4.10	-0.01	1.651	1.274	1.184	0.21434	0.17099	0.22227
158	1	1.253	-4.12	-0.00	1.442	1.355	1.346	0.14474	0.14604	0.09213
158	2	1.253	-2.10	-0.00	1.441	1.305	1.337	0.13422	0.16684	0.09688

TABLE VB
BALANCE CAVITY AND MODEL BASE PRESSURE COEFFICIENTS AND INCREMENTAL
AXIAL FORCE COEFFICIENTS - THRUST EFFECT PHASE

Run	Pt.	M _c	α	φ	C _{Pc}	C _{Pb1}	C _{Pb2}	ΔC _{Ac}	ΔC _{Ab}	C _{Au}
158	3	1.253	-1.06	-0.00	1.415	1.335	1.359	0.13592	0.20208	-0.10738
158	4	1.253	-0.54	-0.00	1.415	1.347	1.345	0.13586	0.20179	-0.110267
158	5	1.253	-0.04	-0.00	1.426	1.359	1.348	0.13962	0.20606	-0.114611
158	6	1.253	0.50	-0.00	1.434	1.345	1.333	0.14213	0.19922	-0.116117
158	7	1.252	1.02	-0.00	1.441	1.328	1.333	0.14751	0.19265	-0.113300
158	8	1.250	3.08	-0.00	1.449	1.267	1.304	0.14756	0.17254	-0.109215
158	9	1.250	4.17	-0.00	1.450	1.268	1.285	0.14831	0.16198	-0.109215
158	10	1.250	4.17	-0.00	1.448	1.235	1.274	0.14769	0.14935	-0.107555
159	1	1.250	4.13	-0.00	1.312	1.228	1.209	0.10288	0.12806	-0.00890
159	2	1.252	-2.11	-0.00	1.309	1.252	1.259	0.10260	0.14919	-0.00890
159	3	1.252	-1.06	-0.00	1.314	1.271	1.271	0.10283	0.15790	-0.022254
159	4	1.252	-0.53	-0.00	1.312	1.269	1.262	0.10229	0.15486	-0.022254
159	5	1.251	-0.09	-0.00	1.314	1.274	1.267	0.10321	0.15603	-0.022331
159	6	1.252	0.08	-0.00	1.316	1.271	1.270	0.10392	0.15822	-0.023331
159	7	1.252	1.08	-0.00	1.313	1.244	1.265	0.10600	0.15776	-0.018950
159	8	1.252	3.13	-0.00	1.319	1.237	1.260	0.10688	0.14527	-0.018950
159	9	1.251	4.17	-0.00	1.312	1.210	1.240	0.10245	0.13154	-0.009543
160	1	1.250	4.12	-0.00	0.769	0.775	0.763	0.07581	0.13490	0.45888
160	2	1.251	-2.12	-0.00	0.776	0.783	0.772	0.07332	0.12966	0.44816
160	3	1.252	-1.07	-0.00	0.780	0.785	0.776	0.07198	0.12791	0.44816
160	4	1.252	-0.54	-0.00	0.781	0.783	0.775	0.07177	0.12843	0.44816
160	5	1.252	0.02	-0.00	0.780	0.782	0.774	0.07193	0.12885	0.44816
160	6	1.252	0.99	-0.00	0.779	0.781	0.774	0.07219	0.12932	0.44816
160	7	1.252	2.09	-0.00	0.777	0.780	0.774	0.07216	0.12967	0.44816
160	8	1.252	3.14	-0.00	0.773	0.779	0.771	0.07291	0.13100	0.45107
160	9	1.251	4.16	-0.00	0.773	0.777	0.767	0.07455	0.13264	0.45556
160	10	1.251	4.16	-0.00	0.766	0.777	0.761	0.07646	0.13463	0.46211
161	3	1.102	1.12	-0.00	0.848	0.852	0.841	0.06418	0.11545	0.41918
161	4	1.100	-2.15	-0.00	0.863	0.863	0.855	0.05853	0.11588	0.41857
161	5	1.102	-1.05	-0.00	0.865	0.864	0.858	0.05775	0.10493	0.40690
161	6	1.103	-0.56	-0.00	0.869	0.867	0.860	0.05705	0.10370	0.40690
161	7	1.103	0.03	-0.00	0.864	0.864	0.854	0.05946	0.10821	0.40627
161	8	1.100	1.08	-0.00	0.865	0.864	0.860	0.05741	0.10408	0.40069
161	9	1.100	2.06	-0.00	0.864	0.862	0.859	0.05685	0.10431	0.40069
161	10	1.101	3.13	-0.00	0.864	0.862	0.858	0.05773	0.10514	0.40811
161	11	1.101	4.13	-0.00	0.857	0.853	0.853	0.06233	0.10823	0.41533
161	12	1.101	4.13	-0.00	0.852	0.853	0.848	0.06268	0.11272	0.42020
161	13	1.101	-0.00	-0.00	0.865	0.863	0.860	0.05717	0.11041	0.42020

TABLE VB
BALANCE CAVITY AND MODEL BASE PRESSURE COEFFICIENTS AND INCREMENTAL AXIAL FORCE COEFFICIENTS - THRUST EFFECT PHASE

Run	Pt.	M _c	α	Φ	C _{Pc}	C _{Pb1}	C _{Pb2}	ΔC _{Ac}	ΔC _{Ab}	C _{Au}
162	1	1.002	-4.11	-0.00	0.873	0.877	0.867	0.06497	0.11620	0.37805
162	2	1.002	-2.17	-0.00	0.877	0.880	0.871	0.06291	0.11285	0.36756
162	3	1.002	-1.56	-0.00	0.878	0.876	0.873	0.06199	0.11432	0.36465
162	4	1.002	-0.53	-0.00	0.880	0.876	0.872	0.06256	0.11250	0.36413
162	5	1.001	0.46	-0.00	0.879	0.879	0.874	0.06151	0.11218	0.36261
162	6	1.001	1.02	-0.00	0.878	0.877	0.874	0.06246	0.11319	0.36301
162	7	1.002	2.07	-0.00	0.875	0.875	0.868	0.06472	0.11580	0.37943
162	8	1.001	3.09	-0.00	0.875	0.870	0.870	0.06397	0.11997	0.39431
162	9	1.003	4.12	-0.00	0.869	0.870	0.864	0.06690	0.11997	0.39431
162	10	0.999	-0.02	-0.00	0.882	0.880	0.875	0.06049	0.11143	0.35714
163	3	1.098	-4.07	0.00	1.509	1.273	1.400	0.21717	0.25514	0.32172
163	4	1.102	-2.11	-0.00	1.512	1.342	1.443	0.21760	0.29729	0.35316
163	5	1.103	-1.06	-0.00	1.505	1.398	1.443	0.21172	0.31673	0.35316
163	6	1.103	0.56	-0.00	1.511	1.430	1.443	0.21063	0.33063	0.37143
163	7	1.102	0.45	-0.00	1.532	1.388	1.418	0.21592	0.33962	0.37143
163	8	1.101	0.98	-0.00	1.541	1.355	1.370	0.22528	0.33884	0.35854
163	9	1.101	0.03	-0.00	1.559	1.310	1.311	0.22963	0.33409	0.35854
163	10	1.099	3.08	-0.00	1.574	1.281	1.311	0.23724	0.34099	0.32228
163	11	1.098	4.11	-0.00	1.585	1.276	1.288	0.24492	0.34679	0.32228
164	1	1.001	-4.05	0.00	1.455	1.310	1.365	0.23360	0.30837	0.3686
164	2	1.000	-2.06	-0.00	1.448	1.366	1.399	0.23139	0.35106	0.45931
164	3	1.000	-1.55	-0.00	1.445	1.392	1.406	0.22875	0.36569	0.46711
164	4	1.001	-0.55	-0.00	1.451	1.409	1.414	0.22879	0.37715	0.47232
164	5	1.001	0.46	-0.00	1.460	1.404	1.414	0.23142	0.37306	0.48046
164	6	1.002	0.98	-0.00	1.480	1.350	1.383	0.24589	0.34506	0.47465
164	7	1.002	0.03	-0.00	1.476	1.315	1.354	0.24469	0.32067	0.45099
164	8	1.000	2.04	-0.00	1.475	1.286	1.321	0.24469	0.29782	0.43301
164	9	1.000	3.12	-0.00	1.488	1.257	1.274	0.25212	0.21695	0.43301
164	10	0.998	-4.11	0.00	1.425	1.258	1.288	0.17506	0.24981	0.38623
165	1	1.000	-2.11	-0.00	1.345	1.284	1.306	0.17639	0.27229	0.30431
165	2	1.002	-1.05	-0.00	1.346	1.311	1.303	0.17639	0.27648	0.30431
165	3	1.000	0.53	-0.00	1.346	1.311	1.306	0.17783	0.28237	0.33166
165	4	1.000	0.46	-0.00	1.361	1.324	1.321	0.18557	0.29454	0.33166
165	5	1.000	0.46	-0.00	1.361	1.324	1.306	0.18667	0.28614	0.33166
165	6	1.000	1.04	-0.00	1.371	1.313	1.301	0.19110	0.28143	0.33166
165	7	1.000	2.00	-0.00	1.371	1.293	1.275	0.19216	0.26143	0.33166
165	8	1.000	3.00	-0.00	1.371	1.293	1.275	0.19216	0.26143	0.33166

TABLE VB
BALANCE CAVITY AND MODEL BASE PRESSURE COEFFICIENTS AND INCREMENTAL
AXIAL FORCE COEFFICIENTS - THRUST EFFECT PHASE

Run	Pt.	M _c	α	Φ	C _{p_c}	C _{p_{b1}}	C _{p_{b2}}	ΔC _{A_c}	ΔC _{A_b}	C _{A_u}
165	9	0.999	3.07	-0.00	1.388	1.268	1.258	0.19994	0.24124	-0.31339
165	10	0.999	4.11	-0.00	1.390	1.253	1.232	0.20131	0.22243	-0.30344
166	1	1.002	-4.10	-0.00	1.266	1.207	1.215	0.13639	0.19243	-0.18294
166	2	1.000	-2.12	-0.00	1.269	1.216	1.216	0.13614	0.19906	-0.19422
166	3	1.000	-1.05	-0.00	1.276	1.231	1.212	0.13906	0.20353	-0.20107
166	4	1.000	-0.56	-0.00	1.279	1.239	1.215	0.14209	0.20627	-0.20663
166	5	0.999	-0.02	-0.00	1.279	1.233	1.217	0.14425	0.20799	-0.20647
166	6	0.998	0.47	-0.00	1.280	1.226	1.221	0.14481	0.20638	-0.20127
166	7	0.999	0.98	-0.00	1.289	1.216	1.219	0.14929	0.19925	-0.19727
166	11	0.999	2.04	-0.00	1.294	1.216	1.210	0.15139	0.19055	-0.19098
166	12	0.999	3.07	-0.00	1.297	1.209	1.199	0.15278	0.17754	-0.18361
166	13	1.000	4.12	-0.00	1.362	1.331	1.320	0.22772	0.29529	-0.47906
167	1	0.902	-4.05	0.00	1.357	1.311	1.322	0.22659	0.34405	-0.57175
167	2	0.901	-2.06	-0.00	1.354	1.313	1.328	0.22544	0.35739	-0.46018
167	3	0.899	-1.06	-0.00	1.362	1.316	1.327	0.23005	0.36890	-0.48618
167	4	0.899	-0.53	-0.00	1.372	1.319	1.328	0.23646	0.33132	-0.46712
167	5	0.900	0.02	-0.00	1.377	1.287	1.271	0.24009	0.30492	-0.48167
167	6	0.899	0.47	-0.00	1.379	1.269	1.260	0.24099	0.28778	-0.46867
167	7	0.900	1.02	-0.00	1.396	1.225	1.222	0.25243	0.25348	-0.45457
167	8	0.900	2.06	-0.00	1.205	1.169	1.165	0.13011	0.18388	-0.20642
167	9	0.898	3.06	-0.00	1.210	1.169	1.169	0.13309	0.18957	-0.21429
167	10	0.898	4.06	-0.00	1.214	1.164	1.169	0.13671	0.19827	-0.22104
168	1	0.900	-4.10	-0.00	1.223	1.183	1.171	0.14126	0.20921	-0.22859
168	2	0.900	-2.08	-0.00	1.227	1.182	1.171	0.14334	0.19984	-0.22501
168	3	0.899	-1.06	-0.00	1.232	1.180	1.171	0.14397	0.19804	-0.22666
168	4	0.900	-0.55	-0.00	1.235	1.172	1.174	0.14741	0.19404	-0.22942
168	5	0.902	0.46	-0.00	1.185	1.152	1.162	0.14928	0.17742	-0.21672
168	6	0.901	1.02	-0.00	1.186	1.145	1.150	0.11759	0.16388	-0.20642
168	7	0.900	2.08	-0.00	1.192	1.151	1.143	0.11827	0.16608	-0.17423
168	8	0.900	3.08	-0.00	1.194	1.158	1.144	0.12072	0.16608	-0.17423
168	9	0.900	4.08	-0.00	1.195	1.157	1.143	0.12331	0.17059	-0.18236
168	10	0.900	-0.03	-0.00	1.195	1.156	1.142	0.12388	0.16844	-0.17997
169	1	0.900	-4.08	0.00	1.185	1.145	1.150	0.11759	0.16388	-0.20642
169	2	0.901	-2.09	-0.00	1.186	1.151	1.143	0.11827	0.16608	-0.17423
169	3	0.900	-1.08	-0.00	1.190	1.158	1.144	0.12072	0.16608	-0.17423
169	4	0.900	-0.55	-0.00	1.192	1.157	1.143	0.12331	0.17059	-0.18236
169	5	0.900	0.03	-0.00	1.194	1.156	1.142	0.12388	0.16844	-0.17997
169	6	0.900	-0.48	-0.00	1.195	1.156	1.142	0.12388	0.16844	-0.17997

TABLE VB
BALANCE CAVITY AND MODEL BASE PRESSURE COEFFICIENTS AND INCREMENTAL
AXIAL FORCE COEFFICIENTS - THRUST EFFECT PHASE

Run	Pt.	M _c	α	φ	C _{Pc}	C _{Pb1}	C _{Pb2}	ΔC _{Ac}	ΔC _{Ab}	C _{Au}
169	7	0.899	0.99	-0.00	1.193	1.154	1.140	0.12297	0.16640	-0.77700
169	8	0.898	2.06	-0.00	1.194	1.148	1.139	0.12359	0.16322	-0.16867
169	9	0.898	3.10	-0.00	1.195	1.143	1.142	0.12471	0.16141	-0.16971
169	10	0.899	4.12	-0.00	1.197	1.131	1.138	0.12572	0.15267	-0.16731
170	1	0.898	-4.11	-0.00	0.929	0.932	0.925	-0.04465	-0.08033	0.27041
170	2	0.901	-2.10	-0.00	0.928	0.929	0.924	-0.04518	-0.08237	0.26628
170	3	0.900	-1.05	-0.00	0.930	0.931	0.926	-0.04403	-0.07958	0.26683
170	4	0.898	-0.54	-0.00	0.930	0.930	0.926	-0.04446	-0.08089	0.26655
170	5	0.900	1.01	-0.00	0.930	0.929	0.925	-0.04432	-0.08170	0.26675
170	6	0.899	0.47	-0.00	0.930	0.929	0.926	-0.04439	-0.08059	0.26663
170	7	0.900	1.02	-0.00	0.929	0.929	0.925	-0.04495	-0.08177	0.26736
170	8	0.901	2.06	-0.00	0.932	0.932	0.928	-0.04290	-0.07845	0.27199
170	9	0.901	3.09	-0.00	0.928	0.930	0.924	-0.04521	-0.08152	0.27599
170	10	0.900	4.11	-0.00	0.925	0.929	0.922	-0.04720	-0.08312	0.27843
171	4	0.798	-2.12	-0.00	1.276	1.218	1.244	0.22314	0.33210	-0.44317
171	5	0.798	-1.04	-0.00	1.275	1.237	1.245	0.22205	0.34598	-0.45341
171	6	0.799	-0.58	-0.00	1.272	1.248	1.250	0.22966	0.35742	-0.46097
171	7	0.800	-0.06	-0.00	1.278	1.242	1.255	0.22358	0.35161	-0.46149
171	8	0.799	0.42	-0.00	1.286	1.216	1.220	0.23048	0.31789	-0.46190
171	9	0.801	1.99	-0.00	1.295	1.204	1.219	0.23618	0.29451	-0.45665
171	10	0.800	3.01	-0.00	1.294	1.185	1.167	0.23618	0.27104	-0.45227
171	11	0.795	4.05	-0.00	1.296	1.171	1.129	0.24058	0.24404	-0.43527
171	12	0.798	-4.09	-0.00	1.311	1.170	1.129	0.25127	0.21547	-0.42782
172	1	0.802	-2.11	-0.00	1.167	1.130	1.135	0.13426	0.18896	-0.21432
172	2	0.799	-1.06	-0.00	1.171	1.139	1.141	0.13818	0.20087	-0.22565
172	3	0.797	-0.59	-0.00	1.175	1.147	1.139	0.13961	0.20356	-0.22880
172	4	0.802	-0.05	-0.00	1.180	1.147	1.141	0.14386	0.20611	-0.22872
172	5	0.803	0.42	-0.00	1.180	1.147	1.141	0.14423	0.20525	-0.22660
172	6	0.797	2.01	-0.00	1.177	1.139	1.136	0.14374	0.19940	-0.21984
172	7	0.802	3.06	-0.00	1.182	1.139	1.135	0.14558	0.19702	-0.21717
172	8	0.802	4.09	-0.00	1.184	1.120	1.124	0.14761	0.19047	-0.21267
172	9	0.798	-4.09	-0.00	1.184	1.120	1.124	0.14891	0.17580	-0.21267
172	10	0.799	-2.09	-0.00	1.122	1.094	1.092	0.09896	0.13399	-0.12227
173	1	0.803	-1.08	-0.00	1.131	1.103	1.094	0.10502	0.14015	-0.12797
173	2	0.802	-0.59	-0.00	1.134	1.106	1.095	0.10756	0.14347	-0.12923
173	3	0.800	-0.05	-0.00	1.132	1.102	1.090	0.10595	0.13783	-0.13051
173	4	0.797	-0.05	-0.00	1.130	1.101	1.090	0.10570	0.13791	-0.12618

TABLE VB
BALANCE CAVITY AND MODEL BASE PRESSURE COEFFICIENTS AND INCREMENTAL
AXIAL FORCE COEFFICIENTS - THRUST EFFECT PHASE

Run	Pt.	M _c	α	Φ	C _{Pc}	C _{Pb1}	C _{Pb2}	ΔC _{Ac}	ΔC _{Ab}	C _{Au}
173	6	0.798	0.41	-0.00	1.131	1.101	1.091	0.10578	0.13795	-0.12473
173	7	0.800	0.96	-0.00	1.130	1.100	1.089	0.10505	0.13554	-0.12320
173	8	0.796	2.01	-0.00	1.126	1.096	1.085	0.10253	0.13107	-0.11720
173	9	0.796	3.08	-0.00	1.127	1.095	1.085	0.10362	0.13067	-0.11720
173	10	0.800	4.07	-0.00	1.128	1.104	1.089	0.10317	0.13141	-0.11571
174	1	0.801	-4.09	-0.00	0.947	0.948	0.944	-0.04192	-0.07586	0.26190
174	2	0.797	-2.11	-0.00	0.946	0.945	0.943	-0.04312	-0.07986	0.26362
174	3	0.798	-1.06	-0.00	0.946	0.947	0.944	-0.04155	-0.07709	0.26604
174	4	0.801	-0.60	-0.00	0.946	0.946	0.944	-0.04242	-0.07728	0.26690
174	5	0.800	-0.04	-0.00	0.948	0.948	0.945	-0.04118	-0.07579	0.26436
174	6	0.798	0.42	-0.00	0.947	0.946	0.945	-0.04236	-0.07763	0.26659
174	7	0.799	0.99	-0.00	0.948	0.946	0.944	-0.04177	-0.07778	0.26428
174	8	0.799	2.04	-0.00	0.947	0.947	0.944	-0.04234	-0.07737	0.26739
174	9	0.800	3.07	-0.00	0.947	0.949	0.944	-0.04218	-0.07601	0.26675
174	10	0.800	4.09	-0.00	0.945	0.948	0.943	-0.04333	-0.07728	0.26695
175	1	0.396	-3.99	-0.00	1.067	1.051	1.057	0.22026	0.31740	-0.38732
175	2	0.397	-2.07	-0.00	1.070	1.055	1.057	0.22811	0.32727	-0.41358
175	3	0.397	-1.05	-0.00	1.069	1.063	1.061	0.22519	0.35271	-0.41557
175	4	0.398	-0.58	-0.00	1.068	1.063	1.064	0.22197	0.36894	-0.42268
175	5	0.400	-0.05	-0.00	1.069	1.063	1.064	0.22581	0.37017	-0.42050
175	6	0.398	0.42	-0.00	1.070	1.061	1.063	0.22738	0.35813	-0.41078
175	7	0.398	0.99	-0.00	1.071	1.056	1.058	0.23168	0.33374	-0.41078
175	8	0.396	2.00	-0.00	1.072	1.051	1.055	0.23827	0.30926	-0.40242
175	9	0.396	3.03	-0.00	1.072	1.049	1.048	0.23730	0.30530	-0.38795
175	10	0.397	4.01	-0.00	1.073	1.047	1.047	0.23859	0.27394	-0.36897
176	1	0.400	-4.03	-0.00	0.989	0.987	0.987	-0.03489	-0.07037	0.26994
176	2	0.401	-2.07	-0.00	0.989	0.987	0.988	-0.03257	-0.06753	0.27688
176	3	0.401	-1.00	-0.00	0.989	0.988	0.988	-0.03304	-0.06639	0.28360
176	4	0.401	-0.60	-0.00	0.989	0.988	0.988	-0.03386	-0.06517	0.28456
176	5	0.401	0.42	-0.00	0.989	0.987	0.988	-0.03271	-0.06760	0.28625
176	6	0.401	0.95	-0.00	0.989	0.987	0.988	-0.03206	-0.06707	0.27781
176	7	0.402	1.98	-0.00	0.989	0.988	0.988	-0.03219	-0.06527	0.27910
176	8	0.401	2.99	-0.00	0.990	0.988	0.988	-0.03086	-0.06688	0.27910
176	9	0.401	4.00	-0.00	0.990	0.988	0.988	-0.03158	-0.06554	0.27860
176	10	0.401	4.00	-0.00	0.990	0.988	0.988	-0.03158	-0.06554	0.27860
178	2	0.800	-4.03	0.00	1.293	1.180	1.223	0.23603	0.28865	-0.46895
178	3	0.800	-2.11	0.00	1.294	1.230	1.249	0.23624	0.34247	-0.49148
178	4	0.798	-1.04	0.00	1.289	1.250	1.253	0.23337	0.36133	-0.50249

TABLE VB
BALANCE CAVITY AND MODEL BASE PRESSURE COEFFICIENTS AND INCREMENTAL
AXIAL FORCE COEFFICIENTS - THRUST EFFECT PHASE

Run	Pt.	M _c	α	Φ	C _{Pc}	C _{Pb1}	C _{Pb2}	ΔC _{Ac}	ΔC _{Ab}	C _{Au}
178	5	0.798	-1.05	0.00	1.291	1.250	1.254	0.23472	0.35238	0.50438
178	6	0.801	-0.54	0.00	1.267	1.266	1.266	0.23002	0.37975	0.50836
178	7	0.800	-0.04	0.00	1.290	1.257	1.267	0.23318	0.37501	0.50873
178	8	0.798	0.45	0.00	1.299	1.231	1.251	0.24166	0.34603	0.50984
178	9	0.800	0.94	0.00	1.309	1.207	1.230	0.24859	0.31295	0.50663
178	10	0.798	2.00	0.00	1.320	1.181	1.190	0.25864	0.26678	0.49783
178	11	0.801	3.00	-0.00	1.327	1.173	1.175	0.26250	0.24853	0.49783
178	12	0.796	4.03	-0.00	1.339	1.177	1.163	0.27573	0.24607	0.47727
179	1	0.800	-4.03	0.00	1.254	1.178	1.198	0.20444	0.26896	0.40545
179	2	0.800	-3.05	0.00	1.254	1.195	1.205	0.20466	0.26658	0.41996
179	3	0.797	-2.06	0.00	1.253	1.209	1.213	0.20469	0.30415	0.42918
179	4	0.797	-1.06	0.00	1.253	1.236	1.232	0.20519	0.33215	0.44388
179	5	0.800	-0.54	0.00	1.253	1.230	1.236	0.20574	0.33766	0.44727
179	6	0.795	-0.06	-0.00	1.253	1.224	1.227	0.20594	0.33544	0.44564
179	7	0.797	0.47	0.00	1.260	1.221	1.223	0.21033	0.32541	0.44602
179	8	0.797	0.96	0.00	1.267	1.213	1.189	0.21656	0.30533	0.43198
179	9	0.798	1.99	-0.00	1.279	1.143	1.151	0.22137	0.26780	0.41465
179	10	0.800	3.04	-0.00	1.286	1.115	1.129	0.23137	0.21499	0.40951
179	11	0.798	4.08	-0.00	1.296	1.113	1.121	0.23816	0.17499	0.40951
179	12	0.797	8.13	-0.00	1.321	1.120	1.109	0.25429	0.17558	0.41999
180	1	0.798	-4.07	0.00	1.170	1.143	1.143	0.13726	0.06022	0.26457
180	2	0.802	-2.07	-0.00	1.183	1.157	1.156	0.14373	0.22272	0.27892
180	3	0.798	-1.05	-0.00	1.181	1.167	1.156	0.14612	0.22472	0.28028
180	4	0.800	-0.55	-0.00	1.185	1.159	1.157	0.14643	0.22341	0.28511
180	5	0.800	0.47	-0.00	1.185	1.155	1.155	0.14837	0.22585	0.28500
180	6	0.803	0.97	-0.00	1.187	1.155	1.156	0.14925	0.22277	0.28665
180	7	0.801	2.01	0.00	1.187	1.150	1.151	0.15038	0.22046	0.28210
180	8	0.801	3.03	-0.00	1.188	1.142	1.144	0.15121	0.21548	0.27365
180	9	0.802	4.08	-0.00	1.190	1.128	1.134	0.15244	0.20687	0.26652
181	1	0.800	-4.05	0.00	1.128	1.105	1.101	0.10315	0.14822	0.17620
181	2	0.798	-3.06	0.00	1.130	1.104	1.098	0.10546	0.14571	0.17648
181	3	0.799	-2.06	0.00	1.133	1.107	1.102	0.10797	0.14920	0.17987
181	4	0.799	-1.06	0.00	1.136	1.109	1.103	0.10930	0.15050	0.17904
181	5	0.800	-0.54	0.00	1.135	1.108	1.104	0.10820	0.15194	0.17939
181	6	0.803	-0.06	0.00	1.131	1.107	1.101	0.10607	0.15100	0.17730
181	7	0.798	0.48	0.00	1.131	1.107	1.102	0.10614	0.15007	0.17492
181	8	0.799	0.97	0.00	1.131	1.107	1.101	0.10614	0.15007	0.17492

TABLE VB
BALANCE CAVITY AND MODEL BASE PRESSURE COEFFICIENTS AND INCREMENTAL
AXIAL FORCE COEFFICIENTS - THRUST EFFECT PHASE

Run	Pt.	M _c	α	φ	C _{Pc}	C _{Pb1}	C _{Pb2}	ΔC _{Ac}	ΔC _{Aa}	ΔC _{Ab}	C _{Au}
181	9	0.803	2.03	0.00	1.131	1.106	1.100	0.10455	0.14730	0.0	0.17052
181	10	0.804	3.02	0.00	1.128	1.105	1.093	0.10212	0.14441	0.0	0.16449
181	11	0.800	4.06	-0.00	1.122	1.100	1.082	0.09851	0.13894	0.0	0.16000
181	12	0.798	6.08	0.00	1.116	1.089	1.075	0.09436	0.12318	0.0	0.14600
181	13	0.799	8.14	0.00	1.112	1.088	1.075	0.09041	0.11714	0.0	0.14142
182	1	0.801	-4.07	0.00	0.939	0.940	0.935	0.04848	0.08810	0.0	0.22121
182	2	0.803	-3.06	0.00	0.941	0.943	0.938	0.04684	0.08356	0.0	0.21695
182	3	0.804	-2.06	0.00	0.942	0.943	0.940	0.04566	0.08166	0.0	0.21425
182	4	0.804	-1.06	0.00	0.943	0.944	0.941	0.04389	0.07867	0.0	0.21205
182	5	0.804	-0.55	0.00	0.944	0.944	0.942	0.04300	0.07804	0.0	0.21239
182	6	0.804	-0.02	0.00	0.944	0.944	0.942	0.04397	0.07860	0.0	0.21438
182	7	0.799	0.48	0.00	0.942	0.943	0.940	0.04512	0.08321	0.0	0.21528
182	8	0.799	1.99	0.00	0.940	0.942	0.938	0.04799	0.08556	0.0	0.21732
182	9	0.799	3.02	0.00	0.940	0.941	0.936	0.04916	0.08825	0.0	0.22141
182	10	0.805	4.03	0.00	0.938	0.939	0.935	0.04924	0.08827	0.0	0.22405
182	11	0.802	6.10	0.00	0.937	0.926	0.926	0.05705	0.10414	0.0	0.24051
182	12	0.801	8.15	0.00	0.928	0.920	0.915	0.06387	0.11676	0.0	0.25536
183	3	0.398	-4.00	0.00	1.072	1.053	1.056	0.23315	0.31866	0.0	0.41807
183	4	0.399	-2.04	0.00	1.073	1.058	1.065	0.23676	0.33074	0.0	0.44503
183	5	0.399	-1.05	0.00	1.072	1.064	1.065	0.23167	0.33074	0.0	0.44480
183	6	0.401	-0.54	0.00	1.073	1.066	1.065	0.23289	0.33233	0.0	0.46291
183	7	0.401	-0.05	0.00	1.074	1.064	1.064	0.23394	0.33681	0.0	0.45291
183	8	0.401	0.45	0.00	1.075	1.060	1.061	0.23851	0.35501	0.0	0.45488
183	9	0.400	1.99	0.00	1.077	1.053	1.056	0.24925	0.33476	0.0	0.43375
183	10	0.398	3.01	0.00	1.077	1.051	1.052	0.25247	0.31476	0.0	0.41609
183	11	0.398	4.00	0.00	1.078	1.050	1.051	0.25582	0.29236	0.0	0.41037
184	1	0.399	-4.00	0.00	1.042	1.035	1.035	0.13750	0.20180	0.0	0.20508
184	2	0.401	-3.09	0.00	1.043	1.036	1.036	0.14132	0.20584	0.0	0.21294
184	3	0.399	-2.06	0.00	1.044	1.035	1.035	0.14009	0.20594	0.0	0.21361
184	4	0.401	-1.05	0.00	1.045	1.036	1.037	0.14358	0.20599	0.0	0.21543
184	5	0.401	-0.04	0.00	1.045	1.036	1.036	0.14550	0.20724	0.0	0.21538
184	6	0.399	0.45	0.00	1.045	1.035	1.036	0.14606	0.20756	0.0	0.21577
184	7	0.401	1.98	0.00	1.044	1.035	1.036	0.14424	0.20855	0.0	0.21412
184	8	0.401	3.98	0.00	1.044	1.034	1.035	0.14298	0.20855	0.0	0.20961
184	9	0.400	5.98	0.00	1.044	1.033	1.034	0.14181	0.19186	0.0	0.19553
184	10	0.400	7.98	0.00	1.044	1.033	1.033	0.14181	0.19186	0.0	0.19553
184	11	0.400	9.98	0.00	1.044	1.033	1.033	0.14181	0.19186	0.0	0.19553

TABLE VB
BALANCE CAVITY AND MODEL BASE PRESSURE COEFFICIENTS AND INCREMENTAL
AXIAL FORCE COEFFICIENTS - THRUST EFFECT PHASE

Run	Pt.	M _c	α	Φ	C _{Pc}	C _{Pb1}	C _{Pb2}	ΔC _{Ac}	ΔC _{Ab}	C _{Au}
184	12	0.399	6.05	0.00	1.044	1.030	1.029	0.14232	0.16999	-0.17926
184	13	0.399	8.05	0.00	1.040	1.027	1.025	0.13155	0.15265	-0.16535
185	1	0.402	-4.00	0.00	0.989	0.986	0.986	0.03300	0.07169	0.23562
185	2	0.403	-3.01	0.00	0.989	0.986	0.988	0.03411	0.06971	0.23375
185	3	0.401	-2.06	0.00	0.989	0.987	0.988	0.03256	0.06759	0.23478
185	4	0.401	-1.04	0.00	0.990	0.987	0.988	0.03184	0.06692	0.23432
185	5	0.401	-0.58	0.00	0.990	0.987	0.988	0.03084	0.06583	0.23505
185	6	0.402	-0.06	-0.00	0.990	0.988	0.988	0.03170	0.06536	0.23588
185	7	0.402	0.45	0.00	0.990	0.988	0.989	0.03076	0.06433	0.23757
185	8	0.402	0.98	0.00	0.990	0.988	0.989	0.03136	0.06411	0.23823
185	9	0.402	1.30	0.00	0.989	0.987	0.987	0.03394	0.06736	0.23983
185	10	0.401	3.00	0.00	0.989	0.987	0.987	0.03403	0.07085	0.24243
185	11	0.401	4.00	0.00	0.989	0.986	0.987	0.03550	0.07350	0.24234
185	12	0.403	6.04	0.00	0.987	0.984	0.985	0.03885	0.08473	0.25573
185	13	0.401	8.07	0.00	0.985	0.984	0.983	0.04594	0.09359	0.27031
186	3	0.897	-4.05	-0.00	0.919	0.921	0.913	0.05108	0.09319	0.22378
186	4	0.901	-3.05	-0.00	0.923	0.925	0.919	0.04858	0.08699	0.22192
186	5	0.900	-2.04	-0.00	0.924	0.927	0.921	0.04788	0.08539	0.22170
186	6	0.902	-1.04	-0.00	0.926	0.929	0.923	0.04627	0.08273	0.22184
186	7	0.899	-0.53	-0.00	0.927	0.926	0.920	0.04867	0.08671	0.22170
186	8	0.903	-0.02	-0.00	0.927	0.923	0.923	0.04597	0.08242	0.22180
186	9	0.901	0.48	-0.00	0.926	0.927	0.922	0.04669	0.08402	0.22180
186	10	0.902	0.98	0.00	0.925	0.927	0.922	0.04704	0.08416	0.22186
186	11	0.899	2.00	0.00	0.924	0.926	0.921	0.04751	0.08515	0.22226
186	12	0.899	3.05	0.00	0.919	0.921	0.914	0.05118	0.09278	0.22255
186	13	0.901	4.09	0.00	0.917	0.919	0.913	0.05240	0.09379	0.22290
186	14	0.901	6.11	0.00	0.903	0.903	0.900	0.06087	0.11037	0.22484
186	15	0.901	8.18	0.00	0.894	0.897	0.887	0.06664	0.12091	0.22613
187	1	0.902	0.65	0.00	1.195	1.167	1.163	0.12152	0.19592	0.23304
187	2	0.899	-3.05	-0.00	1.193	1.172	1.164	0.12422	0.19031	0.23454
187	3	0.900	-2.10	-0.00	1.197	1.169	1.159	0.12279	0.18563	0.23320
187	4	0.901	-1.05	-0.00	1.199	1.170	1.162	0.12592	0.18918	0.23353
187	5	0.901	-0.56	-0.00	1.198	1.168	1.160	0.12592	0.18918	0.23353
187	6	0.900	0.47	-0.00	1.198	1.166	1.159	0.12618	0.18570	0.23367
187	7	0.901	0.98	-0.00	1.198	1.166	1.160	0.12706	0.18383	0.23370
187	8	0.901	1.30	-0.00	1.198	1.162	1.160	0.12560	0.18488	0.23353
187	9	0.901	3.03	-0.00	1.197	1.156	1.155	0.12539	0.17556	0.22569
187	10	0.901	4.06	-0.00	1.196	1.145	1.148	0.12452	0.16538	0.22198

TABLE VB
BALANCE CAVITY AND MODEL BASE PRESSURE COEFFICIENTS AND INCREMENTAL
AXIAL FORCE COEFFICIENTS - THRUST EFFECT PHASE

Run	Pt.	M _c	α	Φ	C _{p_c}	C _{p_{b1}}	C _{p_{b2}}	ΔC _{A_c}	ΔC _{A_b}	C _{A_u}
187	12	0.899	6.09	-0.00	1.186	1.130	1.124	0.11823	0.14442	-0.20028
187	13	0.900	8.17	-0.00	1.180	1.134	1.120	0.11406	0.14350	-0.19878
188	3	0.900	-4.07	0.00	1.223	1.186	1.187	0.14222	0.21125	-0.28399
188	4	0.898	-2.10	0.00	1.230	1.207	1.201	0.14661	0.23109	-0.29817
188	5	0.900	-1.07	0.00	1.234	1.209	1.202	0.14866	0.23244	-0.29846
188	6	0.901	-0.56	0.00	1.236	1.203	1.200	0.14851	0.22853	-0.29716
188	7	0.900	-0.05	0.00	1.235	1.201	1.199	0.14970	0.22702	-0.30005
188	8	0.900	0.48	0.00	1.235	1.201	1.198	0.15042	0.22528	-0.29872
188	9	0.899	0.98	0.00	1.241	1.190	1.197	0.15212	0.22350	-0.29481
188	10	0.899	3.05	0.00	1.242	1.190	1.193	0.15328	0.21646	-0.28874
188	11	0.900	4.04	0.00	1.242	1.186	1.185	0.15383	0.20604	-0.28879
188	12	0.899	4.05	0.00	1.242	1.166	1.172	0.15402	0.19139	-0.27829
189	1	0.899	-4.07	0.00	1.283	1.213	1.232	0.18040	0.25195	-0.37247
189	2	0.902	-3.07	0.00	1.288	1.238	1.247	0.18252	0.27299	-0.38869
189	3	0.902	-2.08	0.00	1.291	1.256	1.258	0.18409	0.28964	-0.39731
189	4	0.900	-1.56	0.00	1.291	1.266	1.264	0.18416	0.29920	-0.39858
189	5	0.900	-0.05	0.00	1.294	1.267	1.263	0.18482	0.29902	-0.40222
189	6	0.900	0.45	-0.00	1.297	1.269	1.263	0.18712	0.30037	-0.40689
189	7	0.901	0.92	0.00	1.305	1.269	1.265	0.18898	0.29182	-0.40473
189	8	0.900	3.05	0.00	1.317	1.257	1.241	0.19308	0.27100	-0.39808
189	9	0.899	4.05	0.00	1.326	1.209	1.220	0.20143	0.24305	-0.37975
189	10	0.900	6.09	0.00	1.333	1.189	1.193	0.21113	0.21601	-0.37885
189	11	0.900	6.15	0.00	1.344	1.195	1.177	0.21768	0.21001	-0.37885
189	12	0.899	-4.06	-0.00	1.344	1.212	1.173	0.21881	0.21774	-0.38686
189	13	0.899	-4.09	0.00	1.369	1.252	1.281	0.22960	0.30172	-0.47378
190	1	0.898	-3.06	0.00	1.360	1.270	1.299	0.22753	0.32081	-0.48355
190	2	0.898	-2.06	0.00	1.360	1.293	1.310	0.22301	0.34222	-0.50359
190	3	0.900	-1.54	0.00	1.355	1.334	1.329	0.22644	0.36948	-0.51747
190	4	0.899	-0.04	0.00	1.355	1.336	1.335	0.22853	0.37848	-0.52257
190	5	0.900	0.47	0.00	1.379	1.313	1.329	0.23853	0.35851	-0.51594
190	6	0.899	0.97	0.00	1.392	1.283	1.250	0.24916	0.32781	-0.51300
190	7	0.899	3.05	0.00	1.399	1.245	1.250	0.25456	0.23435	-0.47840
190	8	0.897	4.06	0.00	1.416	1.207	1.191	0.26446	0.23435	-0.46957
190	9	0.897	6.11	-0.00	1.446	1.156	1.164	0.28288	0.21866	-0.4512
190	10	0.898	6.14	-0.00	1.460	1.156	1.169	0.29288	0.20666	-0.4512

TABLE VB
BALANCE CAVITY AND MODEL BASE PRESSURE COEFFICIENTS AND INCREMENTAL
AXIAL FORCE COEFFICIENTS - THRUST EFFECT PHASE

Run	Pt.	M _c	α	Φ	C _{p_c}	C _{p_{b1}}	C _{p_{b2}}	ΔC _{A_c}	ΔC _{A_b}	C _{A_u}
191	1	0.898	-4.02	0.00	1.380	1.241	1.295	0.24218	0.30364	0.49768
191	2	0.898	-3.04	0.00	1.377	1.269	1.313	0.24022	0.33353	0.51137
191	3	0.901	-2.0A	0.00	1.381	1.303	1.331	0.24141	0.35722	0.52613
191	4	0.898	-1.06	-0.00	1.375	1.332	1.337	0.23803	0.37962	0.53806
191	5	0.900	-0.53	-0.00	1.375	1.335	1.347	0.23935	0.39300	0.54325
191	6	0.898	-0.02	-0.00	1.390	1.301	1.331	0.24864	0.38654	0.54161
191	7	0.898	0.47	-0.00	1.402	1.275	1.308	0.25620	0.35047	0.54094
191	10	0.899	0.99	-0.00	1.420	1.248	1.260	0.26435	0.28770	0.52769
191	11	0.899	1.99	-0.00	1.444	1.236	1.236	0.26793	0.26800	0.50976
191	12	0.899	3.04	-0.00	1.444	1.236	1.215	0.28238	0.25713	0.50997
191	13	0.899	4.05	-0.00	1.444	1.239	1.215	0.28238	0.25713	0.50997
192	4	0.900	-4.07	0.00	1.438	1.257	1.262	0.27806	0.29332	0.53920
192	5	0.898	-1.03	0.00	1.412	1.302	1.340	0.26279	0.35532	0.56900
192	6	0.899	-1.03	0.00	1.407	1.349	1.368	0.25904	0.38692	0.57807
192	7	0.897	-0.05	0.00	1.401	1.349	1.368	0.25616	0.40692	0.58145
192	8	0.897	0.97	0.00	1.438	1.280	1.329	0.27296	0.35011	0.56910
192	9	0.897	2.01	0.00	1.465	1.280	1.297	0.27955	0.32761	0.56917
192	10	0.900	4.05	0.00	1.508	1.281	1.256	0.29556	0.30361	0.55407
192	11	0.902	6.09	-0.00	1.508	1.281	1.240	0.31933	0.32111	0.59756
192	12	0.899	8.16	-0.00	1.508	1.347	1.264	0.32330	0.34536	0.62808
193	3	1.005	-4.08	0.00	1.577	1.354	1.364	0.29402	0.32539	0.57008
193	4	0.999	-2.04	0.00	1.543	1.401	1.406	0.28192	0.36964	0.60352
193	5	0.998	-1.04	0.00	1.527	1.451	1.450	0.27513	0.41151	0.61833
193	6	0.997	-0.00	0.00	1.523	1.464	1.489	0.27193	0.43776	0.61855
193	7	1.000	1.00	0.00	1.568	1.402	1.447	0.28626	0.39011	0.60065
193	8	0.998	2.08	0.00	1.601	1.384	1.350	0.30987	0.36472	0.59159
193	9	0.998	4.10	-0.00	1.651	1.364	1.336	0.33657	0.32772	0.64441
193	10	0.998	6.12	-0.00	1.666	1.448	1.377	0.33625	0.35890	0.64932
193	11	0.999	8.22	-0.00	1.666	1.483	1.377	0.33625	0.39420	0.67932
194	1	0.998	-4.08	0.00	1.476	1.320	1.364	0.24513	0.31314	0.48959
194	2	0.998	-2.04	0.00	1.472	1.384	1.419	0.24390	0.36693	0.52055
194	3	0.999	-1.02	0.00	1.470	1.424	1.429	0.24237	0.38934	0.53367
194	4	0.999	0.02	0.00	1.498	1.350	1.389	0.25719	0.39542	0.53540
194	5	0.998	0.95	-0.00	1.514	1.311	1.332	0.26803	0.34519	0.52207
194	6	0.998	2.09	-0.00	1.544	1.274	1.254	0.28099	0.29407	0.48977
194	7	0.998	4.12	-0.00	1.590	1.248	1.244	0.30270	0.22949	0.51306
194	8	1.001	6.17	-0.00	1.608	1.250	1.253	0.31291	0.25422	0.54463
194	11	0.999	8.17	-0.00	1.608	1.302	1.253	0.31291	0.25422	0.54463

TABLE VB
BALANCE CAVITY AND MODEL BASE PRESSURE COEFFICIENTS AND INCREMENTAL
AXIAL FORCE COEFFICIENTS - THRUST EFFECT PHASE

Run	Pt.	M _c	α	Φ	C _{Pc}	C _{Pb1}	C _{Pb2}	ΔC _{Ac}	ΔC _{Ab}	C _{Au}
195	1	1.000	0.06	0.00	1.365	1.271	1.300	0.18777	0.26138	0.36419
195	2	0.996	-4.06	0.00	1.368	1.307	1.315	0.19060	0.28567	0.36169
195	3	0.998	-2.08	0.00	1.370	1.327	1.335	0.19109	0.30402	0.36822
195	4	1.001	-1.07	0.00	1.371	1.343	1.341	0.19055	0.31217	0.36897
195	5	0.999	-0.56	0.00	1.377	1.342	1.335	0.19339	0.31000	0.39049
195	6	1.000	-0.03	0.00	1.383	1.355	1.335	0.19664	0.31061	0.39052
195	7	1.000	0.46	0.00	1.392	1.356	1.326	0.20164	0.30800	0.39455
195	8	1.000	0.93	0.00	1.403	1.352	1.304	0.21034	0.27727	0.39207
195	9	1.000	3.06	0.00	1.429	1.302	1.281	0.21785	0.25200	0.38440
195	10	1.000	4.08	0.00	1.435	1.271	1.261	0.22390	0.22200	0.37240
195	11	0.999	6.10	0.00	1.444	1.268	1.235	0.22900	0.23054	0.38050
195	12	0.999	8.21	0.00	1.444	1.287	1.222	0.22948	0.23750	0.38744
196	1	1.001	0.02	0.00	1.271	1.226	1.234	0.13931	0.21018	0.24562
196	2	1.002	-3.04	0.00	1.280	1.248	1.245	0.14338	0.22488	0.25733
196	3	0.998	-2.08	0.00	1.289	1.251	1.245	0.14502	0.22799	0.26260
196	4	1.001	-1.04	0.00	1.287	1.253	1.244	0.14716	0.22685	0.26384
196	5	1.001	-0.56	0.00	1.289	1.250	1.242	0.14839	0.22465	0.26515
196	6	1.001	0.46	0.00	1.286	1.242	1.237	0.14820	0.22029	0.26515
196	7	1.001	0.94	0.00	1.294	1.228	1.235	0.15106	0.22198	0.26415
196	8	0.999	3.06	0.00	1.295	1.221	1.234	0.15268	0.22551	0.26577
196	9	0.997	4.06	0.00	1.297	1.221	1.222	0.15335	0.22220	0.26783
196	10	0.999	6.12	0.00	1.292	1.212	1.216	0.15054	0.21960	0.26534
196	11	0.997	8.20	0.00	1.283	1.203	1.189	0.14654	0.18182	0.25330
197	3	0.997	-4.04	0.00	1.166	1.139	1.127	0.08623	0.12709	0.10836
197	4	0.997	-2.09	0.00	1.172	1.144	1.152	0.08916	0.12665	0.11235
197	5	0.998	-1.05	0.00	1.176	1.146	1.154	0.09108	0.12869	0.11170
197	6	1.000	-0.54	0.00	1.177	1.147	1.153	0.09160	0.12892	0.11259
197	7	1.000	0.48	0.00	1.176	1.145	1.152	0.09166	0.12734	0.11065
197	8	0.999	0.93	0.00	1.175	1.144	1.152	0.09105	0.12700	0.10654
197	9	0.998	3.05	0.00	1.172	1.141	1.131	0.08915	0.12681	0.10567
197	10	0.998	4.05	0.00	1.166	1.139	1.131	0.08687	0.12510	0.10205
197	11	0.998	6.12	0.00	1.155	1.131	1.130	0.08540	0.12411	0.10749
197	12	0.999	8.22	0.00	1.147	1.133	1.105	0.07920	0.11941	0.10815

TABLE VB
BALANCE CAVITY AND MODEL BASE PRESSURE COEFFICIENTS AND INCREMENTAL
AXIAL FORCE COEFFICIENTS - THRUST EFFECT PHASE

Run	Pt.	M _c	α	Φ	C _{p c}	C _{p b1}	C _{p b2}	ΔC _{A c}	ΔC _{A b}	C _{A u}
19A	1	0.999	-4.06	-0.00	0.846	0.848	0.836	0.07904	-0.144	0.355
19A	2	0.996	-3.04	0.00	0.853	0.855	0.847	-0.07555	-0.136	0.323
19A	3	0.999	-2.08	0.00	0.855	0.857	0.849	-0.07431	-0.134	0.323
19A	4	0.996	-1.03	0.00	0.856	0.859	0.850	-0.07381	-0.133	0.317
19A	5	0.999	-0.54	0.00	0.859	0.860	0.852	-0.07219	-0.130	0.308
19A	6	0.999	-0.02	0.00	0.860	0.860	0.854	-0.07164	-0.130	0.319
19A	7	1.001	0.48	0.00	0.858	0.857	0.855	-0.07161	-0.131	0.327
19A	8	1.001	0.95	-0.00	0.858	0.857	0.853	-0.07259	-0.132	0.331
19A	9	1.001	3.05	0.00	0.849	0.850	0.843	-0.07776	-0.140	0.338
19A	10	0.997	4.05	0.00	0.843	0.842	0.837	-0.08051	-0.146	0.338
19A	11	0.998	6.12	0.00	0.822	0.819	0.816	-0.09146	-0.167	0.371
19A	12	1.001	8.21	0.00	0.805	0.807	0.795	-0.09979	-0.180	0.399
19A	13	1.001		0.00						
199	1	1.099	-4.03	0.00	1.608	1.296	1.306	0.25875	0.227	0.404
199	2	1.100	-2.07	0.00	1.548	1.371	1.451	0.23282	0.210	0.400
199	3	1.109	-1.02	-0.00	1.551	1.466	1.476	0.23451	0.214	0.409
199	4	1.109	-0.02	-0.00	1.603	1.500	1.500	0.23533	0.216	0.438
199	5	1.100	0.98	-0.00	1.617	1.534	1.437	0.25659	0.208	0.425
199	6	1.101	2.08	-0.00	1.681	1.584	1.379	0.26287	0.204	0.431
199	7	1.096	4.13	-0.00	1.739	1.617	1.316	0.28843	0.200	0.481
199	8	1.097	6.21	-0.00	1.769	1.681	1.287	0.31613	0.204	0.511
199	9	1.097	8.21	-0.00	1.769	1.739	1.306	0.32824	0.204	0.511
200	1	1.093	-4.03	0.00	1.555	1.283	1.315	0.23893	0.228	0.378
200	5	1.099	-2.10	0.00	1.515	1.379	1.445	0.21947	0.211	0.384
200	6	1.096	-1.03	0.00	1.526	1.430	1.455	0.22501	0.206	0.408
200	7	1.096	0.98	0.00	1.572	1.469	1.486	0.22501	0.206	0.424
200	8	1.099	2.03	0.00	1.593	1.564	1.486	0.24513	0.206	0.424
200	9	1.100	4.07	-0.00	1.650	1.631	1.357	0.25273	0.203	0.496
200	10	1.098	6.11	-0.00	1.703	1.703	1.284	0.27612	0.203	0.536
200	11	1.099	8.20	-0.00	1.737	1.737	1.256	0.30134	0.204	0.536
200	12	1.099		-0.00						
201	1	1.100	-4.05	0.00	1.353	1.244	1.276	0.15002	0.196	0.197
201	2	1.097	-2.08	0.00	1.369	1.317	1.324	0.15780	0.191	0.235
201	3	1.100	-1.05	0.00	1.377	1.333	1.336	0.16003	0.187	0.249
201	4	1.100	-0.55	0.00	1.380	1.343	1.339	0.16046	0.187	0.250
201	5	1.097	0.07	0.00	1.383	1.326	1.329	0.16221	0.187	0.252
201	6	1.097	0.49	-0.00	1.401	1.323	1.322	0.16794	0.181	0.253
201	7	1.097	2.01	-0.00	1.411	1.313	1.312	0.17194	0.179	0.252
201	8	1.098	4.01	-0.00	1.411	1.282	1.280	0.17765	0.173	0.240

TABLE VB
BALANCE CAVITY AND MODEL BASE PRESSURE COEFFICIENTS AND INCREMENTAL
AXIAL FORCE COEFFICIENTS - THRUST EFFECT PHASE

Run	Pt.	M _c	α	φ	C _{p c}	C _{p b1}	C _{p b2}	ΔC _{A c}	ΔC _{A b}	C _{A u}
201	9	1.100	3.07	-0.00	1.425	1.248	1.249	0.18081	0.18614	-0.22519
201	10	1.095	4.09	-0.00	1.420	1.218	1.205	0.18055	0.16188	-0.21426
202	1	1.099	-4.06	0.00	0.836	0.838	0.825	-0.06949	-0.12707	0.36661
202	2	1.097	-3.07	0.00	0.845	0.846	0.837	-0.06592	-0.11979	0.35346
202	3	1.097	-2.10	0.00	0.851	0.852	0.843	-0.06337	-0.11535	0.34517
202	4	1.097	-1.05	0.00	0.855	0.856	0.847	-0.06176	-0.11259	0.34334
202	5	1.099	-0.54	0.00	0.851	0.851	0.843	-0.06327	-0.11503	0.34776
202	6	1.099	0.00	0.00	0.853	0.853	0.846	-0.06235	-0.11325	0.34783
202	7	1.098	0.48	0.00	0.849	0.849	0.846	-0.06180	-0.11355	0.34673
202	8	1.098	1.01	0.00	0.849	0.847	0.843	-0.06470	-0.11772	0.34606
202	9	1.098	2.03	0.00	0.848	0.847	0.841	-0.06744	-0.12426	0.35076
202	10	1.099	3.08	0.00	0.841	0.838	0.832	-0.07071	-0.13142	0.36668
202	11	1.099	4.10	0.00	0.833	0.829	0.823	-0.07771	-0.14019	0.41063
202	12	1.100	6.14	0.00	0.796	0.786	0.788	-0.08637	-0.16012	0.43299
202	13	1.099	8.23	0.00	0.773	0.766	0.765	-0.09634	-0.17681	0.43299
203	5	1.249	-2.09	0.00	1.619	1.400	1.431	0.20378	0.24328	-0.32045
203	6	1.248	-1.04	0.00	1.619	1.462	1.453	0.20417	0.27131	-0.34203
203	7	1.250	-0.05	0.00	1.629	1.474	1.556	0.20705	0.30109	-0.35409
203	8	1.248	0.99	0.00	1.668	1.391	1.484	0.22339	0.25684	-0.35191
203	9	1.248	2.03	0.00	1.678	1.350	1.418	0.22353	0.22516	-0.32825
203	10	1.248	4.09	0.00	1.762	1.307	1.346	0.25151	0.19163	-0.33727
203	11	1.248	6.16	-0.00	1.915	1.292	1.309	0.30216	0.17699	-0.40349
203	12	1.248	8.23	-0.00	1.900	1.566	1.353	0.29724	0.26996	-0.43498
204	1	1.249	-4.06	0.00	1.613	1.262	1.304	0.20235	0.16619	-0.27015
204	2	1.249	-2.08	0.00	1.577	1.379	1.419	0.19026	0.23393	-0.28989
204	3	1.250	-1.04	0.00	1.582	1.447	1.451	0.19149	0.26022	-0.30700
204	4	1.250	-0.56	0.00	1.581	1.488	1.470	0.19117	0.28046	-0.31507
204	5	1.249	-0.03	0.00	1.591	1.429	1.507	0.19411	0.25415	-0.31562
204	6	1.249	0.99	0.00	1.617	1.391	1.474	0.20122	0.25330	-0.30814
204	7	1.251	0.49	0.00	1.627	1.362	1.433	0.20945	0.23303	-0.29232
204	8	1.249	2.02	0.00	1.660	1.327	1.390	0.21743	0.20038	-0.28514
204	9	1.249	3.05	0.00	1.704	1.306	1.358	0.21744	0.17671	-0.28514
204	10	1.249	4.09	0.00	1.704	1.307	1.296	0.23204	0.17671	-0.28514
205	1	1.251	-4.08	0.00	1.554	1.292	1.340	0.18261	0.18515	-0.23623
205	2	1.251	-2.10	0.00	1.528	1.396	1.438	0.17366	0.24392	-0.25956
205	3	1.250	-1.06	0.00	1.555	1.450	1.460	0.17886	0.26631	-0.28042
205	4	1.250	-0.04	0.00	1.555	1.443	1.486	0.18278	0.27182	-0.28031
205	5	1.250	0.99	0.00	1.587	1.371	1.419	0.19348	0.23152	-0.28495

TABLE VB
BALANCE CAVITY AND MODEL BASE PRESSURE COEFFICIENTS AND INCREMENTAL
AXIAL FORCE COEFFICIENTS - THRUST EFFECT PHASE

Run	Pt.	M _c	α	φ	C _{Pc}	C _{Pb1}	C _{Pb2}	ΔC _{Ac}	ΔC _{Ab}	C _{Au}
205	6	1.250	2.05	-0.00	1.598	1.342	1.379	0.19695	0.21121	-0.26799
205	7	1.249	4.11	-0.00	1.661	1.254	1.236	0.21786	0.14362	-0.24876
205	8	1.248	6.17	-0.00	1.736	1.201	1.191	0.24293	0.15113	-0.26804
205	9	1.248	8.25	-0.00	1.734	1.386	1.293	0.24240	0.19963	-0.29809
206	3	1.248	-4.06	0.00	1.413	1.267	1.276	0.13639	0.15956	-0.15409
206	4	1.250	-3.06	0.00	1.408	1.318	1.348	0.13420	0.19504	-0.16673
206	5	1.250	-2.09	0.00	1.416	1.349	1.361	0.13699	0.20775	-0.17627
206	6	1.250	-1.05	0.00	1.422	1.375	1.372	0.13893	0.21857	-0.18469
206	7	1.249	-0.56	0.00	1.431	1.375	1.365	0.13926	0.21704	-0.18730
206	8	1.249	-0.46	0.00	1.437	1.368	1.355	0.14184	0.21986	-0.18833
206	9	1.249	0.00	0.00	1.443	1.345	1.358	0.14656	0.20504	-0.18701
206	10	1.249	1.00	0.00	1.457	1.321	1.339	0.15058	0.19319	-0.17988
206	11	1.250	2.09	-0.00	1.447	1.328	1.315	0.15234	0.17673	-0.16501
206	12	1.250	3.09	0.00	1.462	1.288	1.285	0.15179	0.15440	-0.14058
206	13	1.250	4.12	-0.00	1.461	1.269	1.287	0.14842	0.15440	-0.14058
206	14	1.249	6.16	-0.00	1.450	1.259	1.267	0.14679	0.16102	-0.14077
206	15	1.250	8.22	-0.00	1.446	1.276	1.274	0.14679	0.16102	-0.14077
207	1	1.253	4.06	0.00	1.290	1.250	1.239	0.09560	0.14330	-0.07110
207	2	1.251	-3.06	0.00	1.314	1.271	1.266	0.10157	0.15638	-0.08520
207	3	1.251	-2.07	0.00	1.323	1.282	1.268	0.10327	0.15935	-0.09206
207	4	1.251	-1.07	0.00	1.325	1.287	1.264	0.10689	0.15981	-0.09537
207	5	1.249	-0.56	0.00	1.327	1.277	1.262	0.10792	0.15895	-0.09777
207	6	1.249	-0.46	0.00	1.329	1.272	1.255	0.10815	0.15444	-0.09613
207	7	1.249	0.00	0.00	1.328	1.266	1.254	0.10775	0.15214	-0.09559
207	8	1.249	0.99	0.00	1.327	1.260	1.256	0.10748	0.15102	-0.09503
207	9	1.250	2.03	-0.00	1.321	1.253	1.259	0.10648	0.15002	-0.08391
207	10	1.249	3.09	0.00	1.311	1.229	1.247	0.10273	0.13958	-0.07282
207	11	1.249	4.09	0.00	1.289	1.207	1.222	0.09523	0.12582	-0.05022
207	12	1.249	6.16	-0.00	1.278	1.218	1.221	0.09163	0.12591	-0.04804
207	13	1.250	8.25	-0.00	1.278	1.218	1.221	0.09163	0.12591	-0.04804
208	1	1.251	4.09	0.00	0.748	0.748	0.723	-0.08259	0.15411	0.40689
208	2	1.252	-3.08	0.00	0.758	0.760	0.747	-0.07920	0.14344	0.39856
208	3	1.249	-2.08	0.00	0.760	0.761	0.751	-0.07895	0.14286	0.39420
208	4	1.250	-1.03	0.00	0.762	0.763	0.754	-0.07812	0.14087	0.39201
208	5	1.251	-0.56	0.00	0.763	0.763	0.755	-0.07775	0.14034	0.39175
208	6	1.251	-0.46	0.00	0.762	0.762	0.755	-0.07803	0.14085	0.39339
208	7	1.251	0.00	0.00	0.762	0.762	0.754	-0.07792	0.14090	0.39194
208	8	1.252	0.98	0.00	0.759	0.760	0.752	-0.07890	0.14188	0.39651
208	9	1.252	2.07	0.00	0.759	0.760	0.752	-0.07890	0.14188	0.39651

TABLE VB
 BALANCE CAVITY AND MODEL BASE PRESSURE COEFFICIENTS AND INCREMENTAL
 AXIAL FORCE COEFFICIENTS - THRUST EFFECT PHASE

Run Pt.	M _c	α	Φ	C _{Pc}	C _{Pb1}	C _{Pb2}	ΔC _{Ac}	ΔC _{Ab}	C _{Au}
208 10	1.251	3.08	0.00	0.754	0.759	0.747-0.08066	-0.14399	0.40188	
208 11	1.251	4.12	0.00	0.743	0.748	0.737-0.08414	-0.15000	0.41006	
208 12	1.248	6.16	0.00	0.720	0.720	0.715-0.09229	-0.16529	0.43337	
208 13	1.251	8.27	0.00	0.701	0.707	0.696-0.09809	-0.17404	0.44777	

TABLE VB
BALANCE CAVITY AND MODEL BASE PRESSURE COEFFICIENTS AND INCREMENTAL
AXIAL FORCE COEFFICIENTS - THRUST EFFECT PHASE

Run	Pt.	M _c	α	Φ	C _{p,c}	C _{p,b1}	C _{p,b2}	ΔC _{A,c}	ΔC _{A,b}	C _{A,u}
210	3	1.249	-4.06	0.00	1.715	1.270	1.332	0.2596	0.17660	-0.33947
210	4	1.250	-2.09	0.00	1.696	1.490	1.462	0.22918	0.23379	-0.35529
210	5	1.249	-1.04	0.00	1.625	1.515	1.567	0.21410	0.28864	-0.35590
210	6	1.248	-0.96	0.00	1.691	1.431	1.504	0.22618	0.27425	-0.36755
210	7	1.249	2.01	0.00	1.725	1.385	1.434	0.23878	0.23969	-0.35513
210	8	1.250	4.06	0.00	1.747	1.335	1.405	0.24605	0.18168	-0.33304
210	9	1.248	6.14	0.00	1.772	1.375	1.266	0.25459	0.18803	-0.34823
210	11	1.248	8.24	0.00	1.795	1.495	1.369	0.26223	0.25379	-0.40539
211	1	1.250	-4.05	0.00	1.552	1.304	1.332	0.18153	0.18696	-0.23135
211	2	1.250	-2.05	0.00	1.537	1.393	1.419	0.17678	0.23539	-0.26078
211	3	1.250	-1.02	0.00	1.524	1.431	1.455	0.17350	0.26059	-0.27720
211	4	1.250	-0.94	0.00	1.526	1.459	1.463	0.17314	0.27051	-0.27752
211	5	1.250	1.04	0.00	1.566	1.428	1.417	0.18610	0.23262	-0.25396
211	6	1.250	2.11	0.00	1.611	1.395	1.382	0.20107	0.23962	-0.24374
211	7	1.248	4.17	0.00	1.630	1.347	1.392	0.20807	0.21607	-0.22564
211	8	1.249	6.25	0.00	1.669	1.337	1.406	0.22074	0.21807	-0.22564
212	3	1.249	-4.07	0.00	1.361	1.261	1.286	0.11945	0.16070	-0.10221
212	4	1.250	-2.07	0.00	1.365	1.284	1.306	0.12013	0.17574	-0.11049
212	5	1.250	-1.06	0.00	1.361	1.290	1.315	0.11906	0.17645	-0.11373
212	6	1.250	-0.96	0.00	1.360	1.295	1.318	0.11849	0.17837	-0.11368
212	7	1.250	-0.85	0.00	1.363	1.296	1.319	0.11915	0.17751	-0.11097
212	8	1.250	0.07	0.00	1.370	1.301	1.319	0.12440	0.18096	-0.10901
212	9	1.250	1.13	0.00	1.376	1.282	1.305	0.12405	0.17225	-0.10787
212	10	1.250	3.13	0.00	1.383	1.262	1.304	0.12626	0.16555	-0.10409
212	11	1.250	5.28	0.00	1.391	1.234	1.301	0.13358	0.16040	-0.10808
212	11	1.250	8.28	0.00	1.411	1.223	1.301	0.13558	0.15560	-0.10808
213	3	1.250	-4.06	0.00	1.265	1.206	1.209	0.08678	0.22400	-0.05232
213	4	1.250	-2.10	0.00	1.253	1.218	1.211	0.08649	0.23128	-0.05232
213	5	1.250	-1.04	0.00	1.247	1.200	1.207	0.08120	0.22288	-0.04108
213	6	1.250	-0.92	0.00	1.249	1.201	1.207	0.08120	0.21948	-0.04508
213	7	1.250	0.52	0.00	1.251	1.200	1.203	0.08268	0.21916	-0.04371
213	8	1.250	2.48	0.00	1.251	1.202	1.219	0.08279	0.21172	-0.04677
213	9	1.250	4.48	0.00	1.251	1.202	1.219	0.08279	0.21172	-0.04677

TABLE VB
BALANCE CAVITY AND MODEL BASE PRESSURE COEFFICIENTS AND INCREMENTAL AXIAL FORCE COEFFICIENTS - THRUST EFFECT PHASE

Run	Pt.	M_c	α	Φ	C_{p_c}	$C_{p_{b1}}$	$C_{p_{b2}}$	ΔC_{A_c}	ΔC_{A_b}	C_{A_u}
213	9	1.251	2.05	0.00	1.264	1.197	1.228	0.08696	0.12416	0.00777
213	10	1.251	3.08	0.00	1.269	1.169	1.222	0.08856	0.12017	0.00648
213	11	1.250	4.17	0.00	1.264	1.172	1.216	0.08671	0.12021	0.00354
213	13	1.251	6.26	0.00	1.272	1.155	1.193	0.08690	0.10890	0.00205
213	13	1.251	8.26	0.00	1.272	1.155	1.193	0.08959	0.10176	0.00191
214	1	1.250	-4.08	0.00	0.739	0.745	0.718	0.08562	0.15662	0.49293
214	2	1.248	-3.08	0.00	0.747	0.752	0.728	0.08327	0.15212	0.48859
214	3	1.251	-2.10	0.00	0.750	0.755	0.732	0.08189	0.14953	0.48586
214	4	1.252	-1.55	0.00	0.750	0.754	0.732	0.08176	0.14907	0.48234
214	5	1.252	-0.01	0.00	0.749	0.754	0.732	0.08238	0.14978	0.48180
214	6	1.250	0.48	0.00	0.748	0.754	0.732	0.08246	0.14941	0.48354
214	7	1.250	1.04	0.00	0.749	0.754	0.732	0.08227	0.14964	0.48224
214	8	1.249	3.10	0.00	0.746	0.753	0.728	0.08363	0.15148	0.48793
214	10	1.251	4.12	0.00	0.739	0.738	0.722	0.08558	0.15472	0.49348
214	11	1.251	6.19	0.00	0.713	0.730	0.698	0.09395	0.16665	0.53209
214	13	1.251	8.26	0.00	0.702	0.718	0.685	0.09777	0.17377	0.53477
215	1	1.099	-4.08	0.00	0.833	0.834	0.824	0.07064	0.12855	0.4515
215	2	1.099	-3.08	0.00	0.833	0.839	0.830	0.06850	0.12853	0.4539
215	3	1.100	-2.10	0.00	0.843	0.845	0.836	0.06650	0.12953	0.4534
215	4	1.100	-1.55	0.00	0.843	0.842	0.835	0.06536	0.12937	0.4541
215	5	1.100	-0.02	0.00	0.843	0.843	0.835	0.06675	0.12862	0.4536
215	6	1.100	0.49	0.00	0.842	0.830	0.831	0.06851	0.12472	0.4537
215	7	1.100	0.99	0.00	0.842	0.841	0.834	0.06668	0.12244	0.4537
215	8	1.099	2.06	0.00	0.842	0.839	0.831	0.06787	0.12243	0.4530
215	9	1.098	3.13	0.00	0.842	0.822	0.833	0.06687	0.12240	0.4530
215	10	1.098	4.17	0.00	0.800	0.832	0.823	0.07116	0.13004	0.4461
215	11	1.098	6.25	0.00	0.787	0.802	0.789	0.08510	0.15464	0.4810
215	13	1.100	8.25	0.00	0.787	0.793	0.776	0.09010	0.16205	0.5081
216	1	1.099	-4.06	0.00	1.325	1.277	1.266	0.13846	0.86700	0.1645
216	2	1.099	-3.06	0.00	1.330	1.252	1.284	0.14026	0.86700	0.1616
216	3	1.097	-2.10	0.00	1.334	1.273	1.295	0.14280	0.86700	0.1627
216	4	1.097	-1.55	0.00	1.343	1.289	1.296	0.14543	0.86700	0.1619
216	5	1.098	-0.03	0.00	1.343	1.289	1.272	0.14643	0.86700	0.1619
216	6	1.098	0.47	0.00	1.349	1.280	1.268	0.14853	0.86700	0.1610
216	7	1.099	0.99	0.00	1.358	1.287	1.268	0.15253	0.86700	0.1610
216	11	1.100	8.25	0.00	1.358	1.287	1.268	0.15253	0.86700	0.1610

TABLE VB
BALANCE CAVITY AND MODEL BASE PRESSURE COEFFICIENTS AND INCREMENTAL AXIAL FORCE COEFFICIENTS - THRUST EFFECT PHASE

Run	Pt.	M _c	α	φ	C _{p_c}	C _{p_{b1}}	C _{p_{b2}}	ΔC _{A_c}	ΔC _{A_b}	C _{A_u}
216	12	1.098	1.01	-0.00	1.357	1.285	1.255	0.15232	0.20666	-0.20065
216	13	1.100	2.09	-0.00	1.367	1.275	1.254	0.15604	0.20021	-0.19416
216	14	1.100	3.09	-0.00	1.372	1.239	1.229	0.15846	0.17755	-0.18531
216	15	1.101	4.13	-0.00	1.386	1.215	1.210	0.16127	0.16040	-0.16644
216	17	1.101	6.25	-0.00	1.396	1.217	1.185	0.16829	0.15204	-0.16539
217	1	1.094	-4.02	0.00	1.543	3.110	4.09	0.23224	0.27462	-0.39168
217	3	1.092	-2.09	-0.00	1.552	3.211	4.65	0.23558	0.21259	-0.41320
217	4	1.100	-1.04	-0.00	1.535	4.707	4.73	0.22768	0.33374	-0.43167
217	5	1.098	0.96	-0.00	1.581	3.617	4.35	0.25092	0.30601	-0.42067
217	6	1.097	3.07	-0.00	1.601	3.223	4.27	0.25912	0.23091	-0.40155
217	7	1.101	4.11	-0.00	1.610	3.235	4.25	0.26063	0.21704	-0.40199
217	10	1.098	6.23	-0.00	1.688	1.472	3.89	0.22933	0.32706	-0.50599
218	1	1.000	4.08	0.00	0.842	0.842	0.834	0.06095	0.14730	0.23559
218	2	1.000	-2.03	-0.00	0.843	0.846	0.836	0.06231	0.14299	0.23020
218	3	1.000	-1.03	-0.00	0.847	0.846	0.841	0.07845	0.14286	0.22622
218	4	1.000	0.98	-0.00	0.845	0.843	0.838	0.07564	0.14483	0.22714
218	5	1.000	2.09	-0.00	0.838	0.839	0.832	0.08066	0.14488	0.23195
218	6	1.000	4.13	-0.00	0.812	0.813	0.805	0.09635	0.15019	0.23359
218	7	1.001	6.22	-0.00	0.799	0.804	0.794	0.10272	0.14287	0.23889
219	1	1.000	4.08	0.00	1.151	1.116	1.109	0.07749	0.10246	0.05594
219	2	1.000	-2.03	-0.00	1.148	1.117	1.107	0.07667	0.10265	0.05196
219	3	1.000	-1.04	-0.00	1.157	1.117	1.110	0.08079	0.10449	0.04900
219	4	1.000	0.95	-0.00	1.157	1.116	1.105	0.08095	0.10273	0.05053
219	5	1.000	3.07	-0.00	1.157	1.113	1.107	0.08173	0.10072	0.04720
219	6	1.000	4.11	-0.00	1.158	1.115	1.107	0.08133	0.10175	0.04801
219	7	1.000	6.22	-0.00	1.159	1.112	1.106	0.08168	0.10059	0.04722
219	10	1.000	4.09	-0.00	1.151	1.107	1.102	0.07750	0.10817	0.04928
219	11	1.000	6.23	-0.00	1.141	1.106	1.104	0.07728	0.10647	0.04843
219	13	1.000	4.08	-0.00	1.141	1.106	1.104	0.07728	0.10647	0.04843

TABLE VB
BALANCE CAVITY AND MODEL BASE PRESSURE COEFFICIENTS AND INCREMENTAL
AXIAL FORCE COEFFICIENTS - THRUST EFFECT PHASE

Run	Pt.	M _c	α	φ	C _p c	C _p b ₁	C _p b ₂	ΔC _A c	ΔC _A b	C _A u
220	6	0.996	-4.07	-0.00	1.253	1.190	1.207	0.131	0.166	0.204
220	7	0.995	-3.06	-0.00	1.247	1.201	1.213	0.122	0.190	0.207
220	8	0.995	-2.11	-0.00	1.242	1.201	1.208	0.121	0.189	0.201
220	9	0.998	-1.04	-0.00	1.252	1.209	1.203	0.130	0.193	0.196
220	10	0.997	-0.56	-0.00	1.259	1.212	1.204	0.120	0.182	0.190
220	11	0.996	0.48	-0.00	1.263	1.205	1.198	0.134	0.176	0.198
220	12	0.998	0.99	-0.00	1.273	1.202	1.190	0.140	0.181	0.202
220	13	0.998	3.07	-0.00	1.273	1.202	1.190	0.140	0.181	0.202
220	14	0.996	4.10	-0.00	1.273	1.196	1.197	0.141	0.176	0.197
220	15	0.998	6.11	-0.00	1.279	1.167	1.163	0.143	0.169	0.197
220	16	0.998	8.21	-0.00	1.270	1.137	1.125	0.143	0.161	0.188
220	17	0.998		-0.00						
220	18	0.998		-0.00						
221	1	0.998	-4.05	-0.00	1.342	1.260	1.297	0.179	0.256	0.302
221	2	0.997	-3.06	-0.00	1.342	1.260	1.301	0.176	0.266	0.302
221	3	0.995	-2.06	-0.00	1.335	1.260	1.302	0.176	0.270	0.302
221	4	0.999	-1.04	-0.00	1.339	1.252	1.302	0.173	0.274	0.302
221	5	1.000	0.05	-0.00	1.344	1.250	1.296	0.177	0.276	0.302
221	6	0.999	0.48	-0.00	1.347	1.250	1.296	0.178	0.277	0.302
221	7	0.999	0.91	-0.00	1.351	1.250	1.296	0.181	0.278	0.302
221	8	0.999	3.00	-0.00	1.369	1.250	1.296	0.190	0.282	0.302
221	9	0.998	4.09	-0.00	1.379	1.250	1.296	0.195	0.283	0.302
221	10	0.997	6.13	-0.00	1.383	1.252	1.297	0.203	0.287	0.302
221	11	0.997	8.19	-0.00	1.393	1.252	1.297	0.203	0.287	0.302
221	12	0.997		-0.00						
221	13	0.997		-0.00						
222	1	0.993	-4.05	-0.00	1.462	1.315	1.362	0.240	0.342	0.407
222	2	0.996	-3.07	-0.00	1.464	1.315	1.362	0.240	0.342	0.407
222	3	0.997	-2.05	-0.00	1.461	1.315	1.362	0.240	0.342	0.407
222	4	0.997	-1.05	-0.00	1.452	1.315	1.362	0.240	0.342	0.407
222	5	0.999	0.05	-0.00	1.459	1.315	1.362	0.240	0.342	0.407
222	6	0.999	0.48	-0.00	1.469	1.315	1.362	0.240	0.342	0.407
222	7	0.999	0.90	-0.00	1.479	1.315	1.362	0.240	0.342	0.407
222	8	0.999	3.00	-0.00	1.496	1.315	1.362	0.240	0.342	0.407
222	9	0.997	4.07	-0.00	1.496	1.315	1.362	0.240	0.342	0.407
222	10	0.996	6.10	-0.00	1.493	1.315	1.362	0.240	0.342	0.407
222	11	0.999	8.19	-0.00	1.503	1.315	1.362	0.240	0.342	0.407
222	12	0.999		-0.00						
222	13	0.999		-0.00						
223	1	0.997	-4.04	-0.00	1.572	1.321	1.390	0.295	0.326	0.367

TABLE VB
BALANCE CAVITY AND MODEL BASE PRESSURE COEFFICIENTS AND INCREMENTAL
AXIAL FORCE COEFFICIENTS - THRUST EFFECT PHASE

Run	Pt.	M _c	α	Φ	C _{p c}	C _{p b1}	C _{p b2}	ΔC _{A c}	ΔC _{A b}	C _{A u}
2	3	0.997	-2.06	0.00	1.557	1.345	1.419	0.288	0.351	0.603
3	3	0.998	-1.03	0.00	1.522	1.450	1.456	0.277	0.436	0.614
4	3	0.995	0.99	0.00	1.548	1.407	1.416	0.269	0.435	0.610
5	3	1.000	2.05	0.00	1.572	1.379	1.394	0.294	0.535	0.617
6	3	0.996	6.11	0.00	1.586	1.351	1.359	0.303	0.597	0.623
7	3	0.995	8.18	0.00	1.604	1.434	1.327	0.311	0.604	0.627
8	3	0.997	0.05	0.00	1.438	1.294	1.265	0.280	0.271	0.539
9	3	0.999	-2.08	0.00	1.413	1.274	1.336	0.261	0.470	0.563
4	4	0.996	-1.04	0.00	1.493	1.353	1.360	0.250	0.371	0.571
4	4	0.999	0.97	0.00	1.429	1.255	1.360	0.269	0.403	0.571
4	4	0.997	2.00	0.00	1.427	1.290	1.265	0.274	0.370	0.562
4	4	0.996	4.11	0.00	1.455	1.233	1.253	0.280	0.307	0.562
4	4	0.997	6.15	0.00	1.457	1.284	1.232	0.291	0.269	0.571
4	4	0.999	8.15	0.00	1.457	1.312	1.262	0.291	0.254	0.571
5	5	0.997	0.15	0.00	1.346	1.265	1.270	0.224	0.295	0.480
5	5	0.998	-1.03	0.00	1.340	1.205	1.316	0.220	0.351	0.480
5	5	0.997	0.97	0.00	1.356	1.314	1.345	0.217	0.350	0.480
5	5	0.997	2.05	0.00	1.354	1.281	1.241	0.235	0.266	0.480
5	5	0.998	4.10	0.00	1.371	1.295	1.247	0.237	0.277	0.480
5	5	0.997	6.10	0.00	1.365	1.205	1.175	0.235	0.216	0.480
6	6	0.997	0.05	0.00	1.264	1.269	1.235	0.166	0.254	0.353
6	6	0.999	-1.06	0.00	1.261	1.219	1.230	0.170	0.254	0.353
6	6	0.997	0.99	0.00	1.261	1.227	1.235	0.165	0.229	0.353
6	6	0.998	2.05	0.00	1.262	1.225	1.229	0.167	0.250	0.353
6	6	0.998	4.07	0.00	1.273	1.239	1.209	0.169	0.254	0.353
6	6	0.997	6.02	0.00	1.277	1.210	1.162	0.177	0.244	0.353
6	6	0.995	8.05	0.00	1.296	1.202	1.105	0.187	0.207	0.353
6	6	0.997	0.19	0.00	1.293	1.182	1.154	0.187	0.190	0.353

TABLE VB
BALANCE CAVITY AND MODEL BASE PRESSURE COEFFICIENTS AND INCREMENTAL
AXIAL FORCE COEFFICIENTS - THRUST EFFECT PHASE

Run	Pt.	M _c	α	φ	C _{Pc}	C _{Pb1}	C _{Pb2}	ΔC _{Ac}	ΔC _{Ab}	C _{Au}
227	1	0.898	-4.05	-0.00	1.183	1.137	1.144	0.1166	0.1597	0.1954
227	2	0.898	-2.09	-0.00	1.177	1.141	1.140	0.1131	0.1598	0.1872
227	3	0.900	-1.05	-0.00	1.177	1.139	1.137	0.1126	0.1566	0.1801
227	4	0.898	-0.56	-0.00	1.177	1.138	1.130	0.1134	0.1552	0.1773
227	5	0.900	0.46	-0.00	1.180	1.137	1.125	0.1148	0.1512	0.1775
227	6	0.899	0.97	-0.00	1.182	1.136	1.123	0.1158	0.1460	0.1743
227	7	0.899	2.01	-0.00	1.182	1.132	1.120	0.1158	0.1480	0.1735
227	8	0.899	3.04	-0.00	1.182	1.129	1.116	0.1163	0.1439	0.1736
227	9	0.899	4.07	-0.00	1.184	1.128	1.109	0.1177	0.1390	0.1727
227	10	0.897	6.14	-0.00	1.184	1.108	1.106	0.1177	0.1271	0.1638
227	11	0.897	8.20	-0.00	1.184	1.104	1.104	0.1177	0.1087	0.1438
228	1	0.900	-4.08	-0.00	0.925	0.923	0.916	0.0475	0.0943	0.2866
228	2	0.898	-2.06	-0.00	0.925	0.921	0.916	0.0485	0.0912	0.2872
228	3	0.898	-1.04	-0.00	0.925	0.922	0.918	0.0476	0.0902	0.2890
228	4	0.901	-0.56	-0.00	0.927	0.924	0.920	0.0462	0.0874	0.2894
228	5	0.900	0.46	-0.00	0.927	0.924	0.920	0.0462	0.0865	0.2897
228	6	0.897	0.97	-0.00	0.926	0.922	0.920	0.0452	0.0857	0.2890
228	7	0.899	1.01	-0.00	0.924	0.921	0.919	0.0463	0.0851	0.2890
228	8	0.899	2.07	-0.00	0.924	0.921	0.917	0.0463	0.0851	0.2890
228	9	0.898	3.13	-0.00	0.925	0.921	0.916	0.0451	0.0851	0.2890
228	10	0.899	4.13	-0.00	0.922	0.919	0.916	0.0451	0.0851	0.2890
228	11	0.899	6.20	-0.00	0.916	0.914	0.910	0.0523	0.0991	0.2905
228	12	0.899	8.20	-0.00	0.911	0.911	0.903	0.0563	0.1046	0.2907
229	1	0.799	-4.05	-0.00	0.943	0.942	0.938	0.0455	0.0850	0.2818
229	2	0.798	-2.03	-0.00	0.942	0.940	0.936	0.0456	0.0879	0.2827
229	3	0.799	-1.05	-0.00	0.942	0.939	0.937	0.0461	0.0876	0.2849
229	4	0.799	-0.54	-0.00	0.942	0.939	0.936	0.0461	0.0861	0.2855
229	5	0.799	0.47	-0.00	0.942	0.939	0.936	0.0462	0.0861	0.2871
229	6	0.798	0.97	-0.00	0.942	0.939	0.936	0.0463	0.0861	0.2867
229	7	0.798	1.02	-0.00	0.941	0.939	0.936	0.0462	0.0861	0.2867
229	8	0.798	2.06	-0.00	0.941	0.939	0.935	0.0462	0.0861	0.2867
229	9	0.799	3.06	-0.00	0.941	0.939	0.935	0.0462	0.0861	0.2867
229	10	0.799	4.06	-0.00	0.935	0.935	0.927	0.0514	0.0921	0.2867
229	11	0.796	6.19	-0.00	0.932	0.935	0.927	0.0543	0.0982	0.2867
229	12	0.796	8.20	-0.00	0.932	0.935	0.927	0.0543	0.0982	0.2867
230	1	0.796	-4.04	-0.00	1.118	1.086	1.085	0.0962	0.1253	0.1314
230	2	0.799	-3.02	-0.00	1.120	1.091	1.089	0.0971	0.1295	0.1298

TABLE VB
BALANCE CAVITY AND MODEL BASE PRESSURE COEFFICIENTS AND INCREMENTAL
AXIAL FORCE COEFFICIENTS - THRUST EFFECT PHASE

Run	Pt.	M _c	α	φ	C _{Pc}	C _{Pb1}	C _{Pb2}	ΔC _{Ac}	ΔC _{Ab}	C _{Au}	
230	3	0.800	-2.07	0.00	1.122	0.93	1.091	0.9838	0.221	-0.126	73
230	4	0.799	-1.04	0.00	1.123	0.89	1.086	0.9799	0.255	-0.125	77
230	5	0.800	-0.54	0.00	1.122	0.89	1.085	0.9790	0.122	-0.122	98
230	6	0.799	-0.04	0.00	1.122	0.87	1.083	0.9798	0.124	-0.121	82
230	7	0.799	0.46	0.00	1.121	0.86	1.081	0.9772	0.123	-0.121	72
230	8	0.799	0.98	0.00	1.121	0.86	1.082	0.9803	0.120	-0.119	54
230	9	0.798	2.05	0.00	1.122	0.84	1.079	0.9803	0.117	-0.119	50
230	10	0.799	3.06	0.00	1.121	0.84	1.075	0.9777	0.114	-0.118	47
230	11	0.799	4.09	0.00	1.120	0.84	1.075	0.9694	0.106	-0.110	73
230	12	0.797	6.17	0.00	1.117	0.86	1.063	0.9494	0.066	-0.107	38
231	1	0.799	0.05	0.00	1.242	1.78	1.87	1.9503	0.263	-0.374	70
231	2	0.799	-3.02	0.00	1.241	1.81	1.197	1.9505	0.276	-0.399	75
231	3	0.799	-2.09	0.00	1.241	1.81	1.197	1.9430	0.291	-0.407	33
231	4	0.800	-1.05	0.00	1.237	1.43	1.213	1.9183	0.301	-0.410	36
231	5	0.798	-0.55	0.00	1.239	1.19	1.207	1.9435	0.305	-0.409	72
231	6	0.799	-0.03	0.00	1.241	1.07	1.173	1.9082	0.293	-0.403	66
231	7	0.799	0.49	0.00	1.246	1.07	1.173	1.9082	0.293	-0.403	57
231	8	0.799	0.92	0.00	1.251	1.07	1.173	1.9082	0.293	-0.403	57
231	9	0.797	2.03	0.00	1.261	1.36	1.152	1.9082	0.293	-0.403	57
231	10	0.797	3.06	0.00	1.261	1.36	1.152	1.9082	0.293	-0.403	57
231	11	0.797	4.09	0.00	1.261	1.36	1.152	1.9082	0.293	-0.403	57
231	12	0.797	6.15	0.00	1.251	1.72	1.14	1.9082	0.293	-0.403	57
232	1	0.409	-3.99	0.00	0.338	0.30	0.30	1.2647	0.176	-0.179	03
232	2	0.399	-2.06	0.00	0.339	0.31	0.31	1.2526	0.178	-0.179	48
232	3	0.399	-1.04	0.00	0.339	0.30	0.30	1.2643	0.178	-0.179	48
232	4	0.399	-0.54	0.00	0.339	0.30	0.30	1.2643	0.178	-0.179	48
232	5	0.400	-0.04	0.00	0.339	0.30	0.30	1.2643	0.178	-0.179	48
232	6	0.399	0.46	0.00	0.339	0.30	0.30	1.2643	0.178	-0.179	48
232	7	0.399	0.98	0.00	0.339	0.30	0.30	1.2643	0.178	-0.179	48
232	8	0.399	2.05	0.00	0.339	0.30	0.30	1.2643	0.178	-0.179	48
232	9	0.399	3.06	0.00	0.339	0.30	0.30	1.2643	0.178	-0.179	48
232	10	0.399	4.09	0.00	0.339	0.30	0.30	1.2643	0.178	-0.179	48
232	11	0.399	6.10	0.00	0.339	0.30	0.30	1.2643	0.178	-0.179	48
232	12	0.399	8.10	0.00	0.339	0.30	0.30	1.2643	0.178	-0.179	48
233	1	0.403	-4.00	0.00	0.967	0.96	0.96	0.0367	0.075	0.279	69

TABLE VB
BALANCE CAVITY AND MODEL BASE PRESSURE COEFFICIENTS AND INCREMENTAL AXIAL FORCE COEFFICIENTS - THRUST EFFECT PHASE

Run	Pt.	M _c	α	φ	C _{Pc}	C _{Pb1}	C _{Pb2}	ΔC _{Ac}	ΔC _{Ab}	C _{Au}
33	3	0.401	-2.95	0.00	0.986	0.985	0.985	0.04312	-0.08275	0.28445
33	4	0.402	-2.01	0.00	0.987	0.986	0.986	0.03888	-0.07580	0.28547
33	5	0.400	-1.53	0.00	0.987	0.986	0.986	0.03848	-0.07707	0.28549
33	6	0.401	-0.53	0.00	0.987	0.986	0.986	0.04006	-0.07798	0.29230
33	7	0.401	0.45	0.00	0.987	0.986	0.986	0.03914	-0.07632	0.29671
33	8	0.402	1.01	0.00	0.987	0.986	0.986	0.04026	-0.07766	0.29177
33	9	0.402	2.09	0.00	0.987	0.986	0.986	0.03879	-0.07517	0.28343
33	10	0.400	4.03	0.00	0.987	0.985	0.985	0.04141	-0.08174	0.29028
33	11	0.402	6.03	0.00	0.986	0.986	0.986	0.04318	-0.08192	0.28943
33	14	0.401	8.09	0.00	0.986	0.985	0.985	0.04388	-0.08478	0.29796
33	14	0.401	8.09	0.00	0.985	0.984	0.983	0.04614	-0.08888	0.29717
53	3	2.247	-4.14	2.49	1.734	3.04	1.374	0.24276	0.19929	5.5331
53	4	2.238	-1.06	2.50	1.669	4.12	1.477	0.21191	0.25366	5.3403
53	5	2.249	0.00	2.22	1.632	5.07	1.455	0.20750	0.27952	5.3686
53	6	2.249	0.93	2.22	1.676	4.37	1.501	0.22293	0.28018	5.3547
53	8	2.250	1.96	2.22	1.775	4.85	1.446	0.23779	0.26018	5.3481
53	9	2.248	6.06	2.22	1.769	5.25	1.358	0.28656	0.20470	5.4340
53	11	2.249	8.16	2.22	1.909	5.25	1.395	0.29962	0.22118	5.4340
56	1	2.247	-4.17	2.49	3.79	2.52	1.267	0.12542	0.15089	5.3926
56	2	2.250	-2.10	2.22	3.562	2.91	1.307	0.11106	0.17512	5.1173
56	3	2.247	0.96	2.22	3.567	3.04	1.303	0.11714	0.16908	5.1293
56	4	2.247	1.98	2.22	3.500	3.00	1.256	0.12566	0.17769	5.1193
56	6	2.248	6.23	2.22	4.417	2.27	1.375	0.13267	0.17464	5.1014
56	7	2.248	8.23	2.22	4.421	2.27	1.232	0.13053	0.14623	5.1094
56	8	2.251	10.23	2.22	4.421	2.27	1.14	0.13053	0.12703	5.0952
77	1	2.250	-4.10	2.49	0.741	0.749	0.722	0.08355	-0.15759	4.9826
77	2	2.250	-2.10	2.22	0.749	0.751	0.732	0.08336	-0.15129	4.9826
77	3	2.250	0.96	2.22	0.748	0.752	0.732	0.08273	-0.15051	4.9826
77	4	2.249	1.98	2.22	0.748	0.754	0.734	0.08264	-0.14969	4.9826
77	6	2.248	6.23	2.22	0.748	0.751	0.734	0.08640	-0.15049	4.9826
77	7	2.248	8.23	2.22	0.714	0.729	0.725	0.09368	-0.16127	4.9826
77	8	2.250	10.23	2.22	0.714	0.693	0.695	0.09368	-0.17061	4.9826

TABLE VB
BALANCE CAVITY AND MODEL BASE PRESSURE COEFFICIENTS AND INCREMENTAL AXIAL FORCE COEFFICIENTS - THRUST EFFECT PHASE

Run	Pt.	M _c	α	φ	C _{p_c}	C _{p_{b1}}	C _{p_{b2}}	ΔC _{A_c}	ΔC _{A_b}	C _{A_u}
237	9	1.248	8.24	22.49	0.703	0.667	0.683	-0.098	-0.184	0.525
238	4	0.997	-4.17	22.49	0.847	0.843	0.836	-0.080	0.147	0.423
238	5	0.999	-2.11	22.49	0.847	0.846	0.840	-0.078	0.143	0.431
238	6	1.001	-1.09	22.49	0.846	0.844	0.839	-0.078	0.145	0.432
238	7	0.999	0.01	22.49	0.845	0.843	0.838	-0.079	0.145	0.432
238	8	1.001	0.95	22.49	0.846	0.841	0.836	-0.079	0.147	0.432
238	9	0.998	1.05	22.49	0.837	0.841	0.837	-0.080	0.147	0.432
238	10	1.001	6.08	22.49	0.817	0.805	0.809	-0.093	0.175	0.468
238	11	0.996	8.17	22.49	0.803	0.789	0.794	-0.101	0.191	0.473
239	1	0.997	16	50	1.447	1.322	1.325	0.231	0.297	0.733
239	2	0.998	-2.10	50	1.448	1.379	1.368	0.229	0.338	0.489
239	3	0.999	-1.08	50	1.445	1.402	1.403	0.229	0.367	0.501
239	4	0.999	0.08	50	1.466	1.357	1.371	0.224	0.353	0.493
239	5	0.998	1.97	49	1.502	1.319	1.304	0.262	0.286	0.518
239	6	0.995	4.04	49	1.523	1.289	1.227	0.271	0.287	0.478
239	7	1.002	6.13	49	1.510	1.320	1.224	0.260	0.270	0.492
240	1	0.997	16	49	1.251	1.190	1.197	0.129	0.172	0.357
240	2	0.997	-2.12	49	1.244	1.207	1.220	0.127	0.195	0.196
240	3	0.997	-1.01	49	1.258	1.203	1.204	0.133	0.184	0.199
240	4	0.999	0.04	49	1.258	1.204	1.192	0.133	0.181	0.207
240	5	0.996	0.48	49	1.272	1.185	1.190	0.143	0.166	0.194
240	6	0.998	2.02	49	1.286	1.165	1.169	0.143	0.153	0.190
240	7	1.000	4.07	49	1.286	1.157	1.149	0.144	0.155	0.186
240	8	0.998	6.13	49	1.284	1.137	1.111	0.146	0.177	0.180
241	1	1.248	12	50	1.653	1.527	1.492	0.216	0.276	0.715
241	2	0.997	-2.10	50	1.626	1.477	1.451	0.210	0.299	0.327
241	3	1.000	-1.06	50	1.639	1.453	1.426	0.210	0.325	0.350
241	4	0.999	0.01	50	1.633	1.451	1.426	0.210	0.325	0.350
241	5	1.000	0.95	50	1.673	1.401	1.376	0.235	0.270	0.353
241	6	0.998	1.05	50	1.734	1.351	1.329	0.245	0.249	0.310
241	7	1.000	6.08	50	1.768	1.301	1.276	0.255	0.251	0.261
241	8	1.002	8.17	50	1.821	1.249	1.224	0.271	0.270	0.221

TABLE VB
 BALANCE CAVITY AND MODEL BASE PRESSURE COEFFICIENTS AND INCREMENTAL
 AXIAL FORCE COEFFICIENTS - THRUST EFFECT PHASE

Run	Pt.	M _c	α	φ	C _{p c}	C _{p b1}	C _{p b2}	ΔC _{A c}	ΔC _{A b}	C _{A u}
243	1	1.250	-4.16	44.99	1.401	1.253	1.255	0.131	0.140	0.117
243	3	1.250	-2.11	44.99	1.501	1.293	1.306	0.125	0.175	0.114
243	4	1.249	-1.08	44.99	1.563	1.287	1.309	0.122	0.176	0.114
243	5	1.249	0.02	44.99	1.571	1.287	1.289	0.122	0.168	0.107
243	6	1.249	2.07	44.99	1.590	1.271	1.243	0.128	0.163	0.107
243	7	1.249	4.13	45.00	1.415	1.194	1.255	0.141	0.151	0.107
243	8	1.249	6.23	45.00	1.426	1.246	1.252	0.141	0.161	0.120
244	1	1.251	-4.17	44.99	0.750	0.759	0.742	0.079	0.146	0.495
244	2	1.250	-2.11	44.99	0.753	0.747	0.735	0.081	0.166	0.490
244	3	1.250	-1.01	44.99	0.753	0.747	0.732	0.081	0.152	0.485
244	4	1.250	0.97	44.99	0.754	0.745	0.736	0.080	0.147	0.489
244	5	1.251	2.99	44.99	0.754	0.748	0.737	0.080	0.140	0.489
244	6	1.249	4.09	44.99	0.753	0.747	0.735	0.080	0.151	0.488
244	7	1.251	6.14	44.99	0.746	0.731	0.726	0.081	0.157	0.495
244	8	1.250	8.24	44.99	0.700	0.671	0.675	0.092	0.190	0.530
245	1	1.001	-4.11	44.99	0.601	1.700	1.459	0.369	0.103	0.470
245	2	1.001	-2.10	44.99	0.670	1.596	1.381	0.243	0.146	0.496
245	3	1.000	-1.07	44.99	0.700	1.536	1.301	0.230	0.223	0.496
245	4	1.000	0.57	44.99	0.687	1.402	1.139	0.236	0.274	0.496
245	5	1.000	2.07	44.99	0.620	1.302	1.000	0.237	0.222	0.505
245	6	1.000	4.97	44.99	0.622	1.224	0.965	0.247	0.185	0.505
245	7	1.000	7.99	44.99	0.627	1.165	0.941	0.267	0.143	0.507
245	8	1.000	10.36	44.99	0.622	1.120	0.954	0.267	0.107	0.510
245	9	1.000	12.60	44.99	0.622	1.085	0.967	0.268	0.072	0.509
245	10	1.000	15.13	44.99	0.622	1.055	0.953	0.268	0.036	0.509
246	1	1.000	-4.17	44.99	1.253	1.700	1.459	0.369	0.103	0.470
246	2	1.001	-2.11	44.99	1.255	1.596	1.381	0.243	0.146	0.496
246	3	1.000	-1.07	44.99	1.256	1.536	1.301	0.230	0.223	0.496
246	4	1.000	0.57	44.99	1.250	1.402	1.139	0.236	0.274	0.496
246	5	1.000	2.07	44.99	1.253	1.302	1.000	0.237	0.222	0.505
246	6	1.000	4.97	44.99	1.257	1.224	0.965	0.247	0.185	0.505
246	7	1.000	7.99	44.99	1.253	1.165	0.941	0.267	0.143	0.507
246	8	1.000	10.36	44.99	1.252	1.120	0.954	0.267	0.107	0.510
246	9	1.000	12.60	44.99	1.252	1.085	0.967	0.268	0.072	0.509
246	10	1.000	15.13	44.99	1.250	1.055	0.953	0.268	0.036	0.509

TABLE VB
BALANCE CAVITY AND MODEL BASE PRESSURE COEFFICIENTS AND INCREMENTAL AXIAL FORCE COEFFICIENTS - THRUST EFFECT PHASE

Run	Pt.	M _c	α	Φ	C _{Pc}	C _{Pb1}	C _{Pb2}	ΔC _{Ac}	ΔC _{Ab}	C _{Au}
246	6	0.996	0.03	44.99	1.258	1.208	1.205	0.13368	0.19064	0.20542
246	7	0.995	0.495	44.99	1.260	1.203	1.205	0.13529	0.18660	0.20400
246	8	0.996	1.96	44.99	1.257	1.203	1.205	0.13551	0.18671	0.20221
246	9	0.997	3.02	44.99	1.275	1.164	1.197	0.13812	0.17482	0.19823
246	10	0.997	4.02	44.99	1.282	1.146	1.190	0.14232	0.16319	0.19436
246	11	0.997	6.06	45.00	1.291	1.137	1.178	0.14580	0.14960	0.19015
246	12	0.995	8.14	45.00	1.286	1.153	1.170	0.14883	0.14920	0.21594
247	1	0.998	-4.20	44.99	0.838	0.838	0.829	0.06323	0.15255	0.43842
247	2	0.999	-3.13	44.99	0.841	0.841	0.834	0.06152	0.14867	0.43555
247	3	0.995	-1.18	44.99	0.851	0.845	0.843	0.07601	0.14206	0.43835
247	4	0.996	-0.58	44.99	0.849	0.845	0.841	0.07801	0.14366	0.43851
247	5	0.998	0.00	44.99	0.846	0.844	0.839	0.07905	0.14522	0.43567
247	6	0.997	0.23	44.99	0.845	0.842	0.838	0.07929	0.14692	0.43570
247	7	0.998	1.97	44.99	0.843	0.838	0.835	0.08068	0.14610	0.43573
247	8	0.999	3.01	44.99	0.841	0.835	0.829	0.08153	0.15175	0.44010
247	9	0.997	4.08	44.99	0.838	0.830	0.824	0.08342	0.16103	0.44340
247	10	0.996	6.15	45.00	0.813	0.802	0.776	0.09680	0.18103	0.46868
247	11	0.997	8.15	45.00	0.789	0.770	0.770	0.10880	0.20103	0.48668

TABLE VB
BALANCE CAVITY AND MODEL BASE PRESSURE COEFFICIENTS AND INCREMENTAL
AXIAL FORCE COEFFICIENTS - THRUST EFFECT PHASE

Run	Pt.	M _c	α	φ	C _{Pc}	C _{Pb1}	C _{Pb2}	ΔC _{Ac}	ΔC _{Ab}	C _{Au}
914	3	0.909	4.03	0.00	1.379	1.44	1.265	0.240	0.296	0.793
914	4	0.899	-3.01	0.00	1.381	1.27	1.303	0.222	0.316	0.494
914	5	0.899	-2.10	0.00	1.368	1.32	1.334	0.234	0.233	0.508
914	6	0.900	-1.06	0.00	1.365	1.22	1.342	0.242	0.371	0.485
914	7	0.900	0.97	0.00	1.368	1.25	1.289	0.249	0.384	0.514
914	8	0.899	2.02	0.00	1.396	1.22	1.247	0.253	0.285	0.501
914	9	0.899	3.05	0.00	1.404	1.24	1.211	0.266	0.444	0.485
914	10	0.899	4.07	0.00	1.437	1.23	1.206	0.270	0.258	0.517
914	11	0.901	6.16	0.00	1.437	1.20	1.218	0.276	0.401	0.514
914	12	0.898	12.29	0.00	1.428	1.29	1.183	0.271	0.289	0.480
915	1	0.995	0.05	0.00	1.470	1.15	1.353	0.243	0.308	0.712
915	2	0.998	-4.07	0.00	1.476	1.30	1.369	0.243	0.260	0.476
915	3	0.999	-2.02	0.00	1.466	1.36	1.409	0.241	0.356	0.494
915	4	0.998	-1.02	0.00	1.456	1.42	1.426	0.240	0.381	0.510
915	5	0.999	0.93	0.00	1.453	1.47	1.370	0.235	0.389	0.512
915	6	0.999	2.06	0.00	1.502	1.20	1.266	0.234	0.282	0.481
915	7	0.999	3.06	0.00	1.520	1.22	1.266	0.235	0.268	0.483
915	8	0.999	4.02	0.00	1.544	1.24	1.266	0.235	0.254	0.483
915	9	0.999	6.23	0.00	1.536	1.21	1.221	0.235	0.270	0.490
915	10	0.999	10.34	0.00	1.536	1.21	1.221	0.235	0.270	0.490
915	11	0.999	12.34	0.00	1.536	1.21	1.221	0.235	0.270	0.490
916	5	0.951	4.03	0.00	1.598	1.42	1.303	0.285	0.278	0.733
916	6	0.947	-3.02	0.00	1.463	1.27	1.363	0.266	0.345	0.502
916	7	0.944	-1.02	0.00	1.440	1.36	1.392	0.264	0.387	0.510
916	8	0.946	0.97	0.00	1.436	1.41	1.354	0.265	0.412	0.507
916	9	0.945	2.06	0.00	1.496	1.25	1.285	0.269	0.295	0.505
916	10	0.945	3.06	0.00	1.512	1.27	1.294	0.269	0.280	0.505
916	11	0.945	4.06	0.00	1.522	1.23	1.291	0.269	0.267	0.505
916	12	0.945	12.26	0.00	1.521	1.21	1.242	0.269	0.331	0.505

TABLE VB
BALANCE CAVITY AND MODEL BASE PRESSURE COEFFICIENTS AND INCREMENTAL
AXIAL FORCE COEFFICIENTS - THRUST EFFECT PHASE

Run	Pt.	M _c	α	Φ	C _{Pc}	C _{Pb1}	C _{Pb2}	ΔC _{Ac}	ΔC _{Ab}	C _{Au}
917	1	048	-4.06	0.00	1.524	1.324	1.395	0.245	0.304	0.438
917	2	053	-3.04	0.00	1.528	1.340	1.420	0.241	0.313	0.441
917	3	049	-2.04	0.00	1.518	1.380	1.446	0.241	0.342	0.453
917	4	048	-1.04	0.00	1.506	1.430	1.461	0.236	0.379	0.463
917	5	049	-0.03	0.00	1.517	1.497	1.498	0.234	0.420	0.474
917	10	051	1.00	0.00	1.561	1.349	1.327	0.251	0.404	0.459
917	11	049	2.04	0.00	1.570	1.366	1.308	0.267	0.457	0.491
917	12	049	3.06	0.00	1.571	1.349	1.308	0.266	0.499	0.517
917	13	054	4.12	0.00	1.593	1.375	1.308	0.272	0.546	0.577
917	14	049	6.19	0.00	1.593	1.428	1.340	0.277	0.600	0.633
917	15	054	8.30	0.00	1.593	1.452	1.340	0.277	0.656	0.690
917	16	054	10.36	0.00	1.603	1.441	1.327	0.279	0.718	0.737
917	17	054	12.36	0.00	1.603	1.412	1.277	0.279	0.780	0.782
918	1	070	-4.05	0.00	1.446	1.294	1.359	0.243	0.379	0.501
918	2	071	-3.07	0.00	1.447	1.326	1.368	0.240	0.403	0.515
918	3	070	-2.04	0.00	1.431	1.346	1.378	0.235	0.427	0.528
918	4	069	-1.02	0.00	1.425	1.363	1.394	0.232	0.451	0.541
918	5	072	0.23	0.00	1.451	1.357	1.392	0.232	0.476	0.554
918	10	073	2.05	0.00	1.462	1.329	1.292	0.253	0.517	0.592
918	11	073	3.05	0.00	1.462	1.302	1.255	0.259	0.562	0.625
918	12	071	4.07	0.00	1.466	1.261	1.244	0.259	0.601	0.657
918	13	074	6.17	0.00	1.472	1.201	1.202	0.255	0.643	0.703
918	14	075	8.25	0.00	1.480	1.111	1.262	0.258	0.685	0.751
918	15	074	10.33	0.00	1.480	1.011	1.202	0.258	0.729	0.791
918	16	074	12.33	0.00	1.480	0.911	1.202	0.258	0.773	0.821
919	1	047	-4.05	0.00	1.409	1.271	1.069	0.234	0.366	0.494
919	2	047	-3.07	0.00	1.417	1.287	1.069	0.233	0.390	0.510
919	3	047	-2.05	0.00	1.407	1.317	1.069	0.232	0.414	0.527
919	4	050	-1.04	0.00	1.398	1.351	1.069	0.232	0.438	0.544
919	5	050	0.99	0.00	1.435	1.366	1.069	0.232	0.462	0.561
919	6	046	2.00	0.00	1.435	1.326	1.069	0.232	0.486	0.578
919	7	046	3.00	0.00	1.435	1.266	1.069	0.232	0.510	0.595
919	8	046	4.00	0.00	1.435	1.166	1.069	0.232	0.534	0.612
919	9	046	6.00	0.00	1.435	1.066	1.069	0.232	0.558	0.629
919	10	046	8.00	0.00	1.435	0.966	1.069	0.232	0.582	0.646
919	11	046	10.00	0.00	1.435	0.866	1.069	0.232	0.606	0.663
919	12	046	12.00	0.00	1.435	0.766	1.069	0.232	0.630	0.680
919	13	046	14.00	0.00	1.435	0.666	1.069	0.232	0.654	0.697
919	14	046	16.00	0.00	1.435	0.566	1.069	0.232	0.678	0.714
919	15	046	18.00	0.00	1.435	0.466	1.069	0.232	0.702	0.731
919	16	046	20.00	0.00	1.435	0.366	1.069	0.232	0.726	0.748
919	17	046	22.00	0.00	1.435	0.266	1.069	0.232	0.750	0.765
919	18	046	24.00	0.00	1.435	0.166	1.069	0.232	0.774	0.782
919	19	046	26.00	0.00	1.435	0.066	1.069	0.232	0.798	0.799
919	20	046	28.00	0.00	1.435	0.066	1.069	0.232	0.822	0.816
919	21	046	30.00	0.00	1.435	0.066	1.069	0.232	0.846	0.833
919	22	046	32.00	0.00	1.435	0.066	1.069	0.232	0.870	0.850
919	23	046	34.00	0.00	1.435	0.066	1.069	0.232	0.894	0.867
919	24	046	36.00	0.00	1.435	0.066	1.069	0.232	0.918	0.884
919	25	046	38.00	0.00	1.435	0.066	1.069	0.232	0.942	0.901
919	26	046	40.00	0.00	1.435	0.066	1.069	0.232	0.966	0.918
919	27	046	42.00	0.00	1.435	0.066	1.069	0.232	0.990	0.935
919	28	046	44.00	0.00	1.435	0.066	1.069	0.232	1.014	0.952
919	29	046	46.00	0.00	1.435	0.066	1.069	0.232	1.038	0.969
919	30	046	48.00	0.00	1.435	0.066	1.069	0.232	1.062	0.986
919	31	046	50.00	0.00	1.435	0.066	1.069	0.232	1.086	1.003
919	32	046	52.00	0.00	1.435	0.066	1.069	0.232	1.110	1.020
919	33	046	54.00	0.00	1.435	0.066	1.069	0.232	1.134	1.037
919	34	046	56.00	0.00	1.435	0.066	1.069	0.232	1.158	1.054
919	35	046	58.00	0.00	1.435	0.066	1.069	0.232	1.182	1.071
919	36	046	60.00	0.00	1.435	0.066	1.069	0.232	1.206	1.088
919	37	046	62.00	0.00	1.435	0.066	1.069	0.232	1.230	1.105
919	38	046	64.00	0.00	1.435	0.066	1.069	0.232	1.254	1.122
919	39	046	66.00	0.00	1.435	0.066	1.069	0.232	1.278	1.139
919	40	046	68.00	0.00	1.435	0.066	1.069	0.232	1.302	1.156
919	41	046	70.00	0.00	1.435	0.066	1.069	0.232	1.326	1.173
919	42	046	72.00	0.00	1.435	0.066	1.069	0.232	1.350	1.190
919	43	046	74.00	0.00	1.435	0.066	1.069	0.232	1.374	1.207
919	44	046	76.00	0.00	1.435	0.066	1.069	0.232	1.398	1.224
919	45	046	78.00	0.00	1.435	0.066	1.069	0.232	1.422	1.241
919	46	046	80.00	0.00	1.435	0.066	1.069	0.232	1.446	1.258
919	47	046	82.00	0.00	1.435	0.066	1.069	0.232	1.470	1.275
919	48	046	84.00	0.00	1.435	0.066	1.069	0.232	1.494	1.292
919	49	046	86.00	0.00	1.435	0.066	1.069	0.232	1.518	1.309
919	50	046	88.00	0.00	1.435	0.066	1.069	0.232	1.542	1.326
919	51	046	90.00	0.00	1.435	0.066	1.069	0.232	1.566	1.343
919	52	046	92.00	0.00	1.435	0.066	1.069	0.232	1.590	1.360
919	53	046	94.00	0.00	1.435	0.066	1.069	0.232	1.614	1.377
919	54	046	96.00	0.00	1.435	0.066	1.069	0.232	1.638	1.394
919	55	046	98.00	0.00	1.435	0.066	1.069	0.232	1.662	1.411
919	56	046	100.00	0.00	1.435	0.066	1.069	0.232	1.686	1.428

TABLE VB
BALANCE CAVITY AND MODEL BASE PRESSURE COEFFICIENTS AND INCREMENTAL AXIAL FORCE COEFFICIENTS - THRUST EFFECT PHASE

Run	Pt.	M _c	α	Φ	C _{Pc}	C _{Pb1}	C _{Pb2}	ΔC _{Ac}	ΔC _{Ab}	C _{Au}
920	1	0.949	-4.06	-0.00	320	1.242	1.256	0.18530	0.25302	-0.38045
920	2	0.948	-3.067	-0.00	310	1.267	1.277	0.18295	0.26752	-0.38219
920	3	0.946	-2.005	-0.00	312	1.273	1.280	0.18214	0.27240	-0.38490
920	4	0.944	-1.002	-0.00	314	1.276	1.284	0.18019	0.28430	-0.38565
920	5	0.947	1.004	-0.00	325	1.262	1.269	0.18601	0.27067	-0.38276
920	6	0.946	3.006	-0.00	343	1.229	1.234	0.20255	0.26661	-0.38224
920	7	0.947	4.006	-0.00	353	1.235	1.236	0.20647	0.22090	-0.37495
920	8	0.945	6.018	-0.00	353	1.234	1.219	0.20647	0.23490	-0.37323
920	9	0.946	10.016	-0.00	330	1.236	1.179	0.20995	0.22124	-0.37420
920	10	0.947	12.034	-0.00	298	1.196	1.110	0.17111	0.15604	-0.28122
920	11	0.947								
920	12	0.947								
920	13	0.947								
921	1	1.050	7.07	0.00	310	1.240	1.257	0.14510	0.20607	-0.17985
921	2	1.045	3.009	-0.00	310	1.251	1.262	0.14583	0.21506	-0.19058
921	3	1.047	-2.007	-0.00	314	1.267	1.265	0.14679	0.21686	-0.19314
921	4	1.050	-1.003	-0.00	329	1.264	1.260	0.15180	0.21680	-0.19819
921	5	1.050	0.006	-0.00	344	1.256	1.252	0.15063	0.21395	-0.19525
921	6	1.050	3.007	-0.00	345	1.251	1.242	0.16087	0.21222	-0.19264
921	7	1.051	4.008	-0.00	356	1.221	1.207	0.16476	0.21035	-0.18592
921	8	1.050	6.013	-0.00	326	1.211	1.189	0.16654	0.17719	-0.16899
921	9	1.050	10.030	-0.00	304	1.167	1.092	0.16654	0.16203	-0.14905
921	10	1.050	12.041	-0.00	264	1.119	1.092	0.13246	0.12037	-0.09052
921	11	1.050								
921	12	1.050								
921	13	1.050								
922	1	1.052	8.007	0.00	202	1.159	1.164	0.09660	0.13298	-0.04132
922	2	1.046	3.009	-0.00	193	1.152	1.153	0.09501	0.13095	-0.04104
922	3	1.047	-2.004	-0.00	191	1.152	1.143	0.09108	0.12335	-0.02297
922	4	1.045	-1.002	-0.00	201	1.152	1.143	0.09494	0.12461	-0.02355
922	5	1.045	0.001	-0.00	219	1.129	1.142	0.09694	0.12270	-0.02292
922	6	1.045	3.008	-0.00	220	1.129	1.142	0.09553	0.12067	-0.02340
922	7	1.045	4.009	-0.00	205	1.115	1.114	0.10143	0.12099	-0.02490
922	8	1.045	6.013	-0.00	173	1.111	1.097	0.09687	0.10910	-0.02162
922	9	1.045	10.030	-0.00	160	1.101	1.029	0.09687	0.10251	-0.01622
922	10	1.045	12.041	-0.00	110	1.032	1.029	0.08169	0.09163	-0.00712
922	11	1.045								
922	12	1.045								
922	13	1.045								

TABLE VB
BALANCE CAVITY AND MODEL BASE PRESSURE COEFFICIENTS AND INCREMENTAL
AXIAL FORCE COEFFICIENTS - THRUST EFFECT PHASE

Run	Pt.	M _c	α	φ	C _{Pc}	C _{Pb1}	C _{Pb2}	ΔC _{Ac}	ΔC _{Ab}	C _{Au}
923	3	1.098	-3.07	0.00	1.244	1.185	1.197	0.10431	0.14515	0.07834
923	4	1.099	-2.04	-0.00	1.245	1.187	1.198	0.10319	0.14774	0.07468
923	5	1.100	-1.02	-0.00	1.253	1.187	1.165	0.10735	0.13246	0.06734
923	6	1.101	0.03	-0.00	1.258	1.183	1.176	0.10746	0.13743	0.06736
923	7	1.109	2.10	-0.00	1.264	1.189	1.174	0.10946	0.13177	0.06298
923	8	1.106	3.11	-0.00	1.256	1.190	1.174	0.11248	0.13865	0.07752
923	9	1.100	4.15	-0.00	1.259	1.163	1.162	0.10887	0.12504	0.05659
923	10	1.109	6.23	-0.00	1.254	1.143	1.137	0.10497	0.10475	0.02981
923	11	1.100	8.34	-0.00	1.256	1.137	1.120	0.10888	0.10967	0.02859
923	12	1.109	10.41	-0.00	1.252	1.107	1.039	0.10728	0.10771	0.02859
923	13	1.102	4.10	0.00	1.110	1.075	0.542	0.04673	0.04879	0.11330
923	14	1.100	3.10	0.00	1.107	1.073	0.623	0.04576	0.05209	0.11659
923	15	1.100	2.05	-0.00	1.108	1.069	0.622	0.04608	0.05179	0.11934
923	16	1.102	-0.01	-0.00	1.103	1.061	0.563	0.04333	0.04502	0.12265
923	17	1.103	1.04	-0.00	1.101	1.058	0.644	0.04296	0.04720	0.12510
923	18	1.102	3.08	-0.00	1.109	1.059	0.644	0.04244	0.04672	0.12830
923	19	1.103	4.15	-0.00	1.100	1.048	0.645	0.04297	0.04724	0.13099
923	20	1.104	6.23	-0.00	1.085	1.040	0.511	0.03593	0.03757	0.13054
923	21	1.105	8.34	0.00	1.073	1.013	0.420	0.03141	0.03077	0.13054
923	22	1.105	10.46	0.00	1.063	0.992	0.287	0.02660	0.02747	0.13712
925	1	1.250	4.09	-0.00	1.110	1.081	0.727	0.03618	0.04483	0.15067
925	2	1.251	3.09	-0.00	1.112	1.079	0.673	0.03570	0.04296	0.15467
925	3	1.252	2.05	-0.00	1.110	1.077	0.669	0.03660	0.04377	0.15179
925	4	1.252	1.02	-0.00	1.111	1.070	0.668	0.03667	0.04280	0.15472
925	5	1.252	0.99	-0.00	1.111	1.065	0.674	0.03638	0.04208	0.15675
925	6	1.252	2.13	-0.00	1.113	1.061	0.719	0.03666	0.04059	0.15828
925	7	1.252	3.16	-0.00	1.107	1.060	0.633	0.03505	0.03849	0.16521
925	8	1.254	4.29	-0.00	1.107	1.046	0.539	0.03507	0.03507	0.17249
925	9	1.254	6.46	-0.00	1.107	1.022	0.333	0.03372	0.02618	0.17155
925	10	1.254	8.46	-0.00	1.113	1.002	0.220	0.03372	0.02618	0.17155
926	3	1.051	-4.07	-0.00	1.086	1.050	1.024	0.04016	0.03106	0.12817

TABLE VB
BALANCE CAVITY AND MODEL BASE PRESSURE COEFFICIENTS AND INCREMENTAL
AXIAL FORCE COEFFICIENTS - THRUST EFFECT PHASE

Run	Pt.	M _c	α	φ	C _{Pc}	C _{Pb1}	C _{Pb2}	ΔC _{Ac}	ΔC _{Ab}	C _{Au}
926	4	1.051	-3.06	-0.00	1.081	1.047	1.025	0.03806	0.03032	0.13903
926	5	1.045	-2.08	-0.00	1.068	1.036	1.015	0.03207	0.02162	0.13660
926	6	1.050	-1.03	-0.00	1.080	1.050	1.029	0.03903	0.03519	0.14469
926	7	1.050	-1.00	-0.00	1.078	1.042	1.030	0.03756	0.03120	0.14688
926	8	1.052	3.06	-0.00	1.074	1.038	1.028	0.03663	0.02787	0.15124
926	9	1.042	4.10	-0.00	1.070	1.044	1.035	0.03625	0.03277	0.14977
926	10	1.051	6.13	-0.00	1.065	1.041	1.031	0.03290	0.03202	0.15943
926	11	1.053	6.23	-0.00	1.060	1.039	1.023	0.03097	0.02876	0.15943
926	12	1.051	10.41	-0.00	1.049	1.033	1.002	0.02275	0.02738	0.16596
926	13	1.051	12.41	-0.00	1.026	1.015	0.951	0.01246	0.02095	0.19644
927	1	1.047	-4.09	-0.00	0.828	0.819	0.808	-0.08414	-0.15511	0.49396
927	2	1.046	-3.08	-0.00	0.834	0.823	0.818	-0.08061	-0.14846	0.49605
927	3	1.045	-2.05	-0.00	0.843	0.831	0.825	-0.07706	-0.14167	0.47600
927	4	1.047	-1.02	-0.00	0.852	0.823	0.814	-0.06185	-0.15285	0.50235
927	5	1.049	1.04	-0.00	0.834	0.830	0.823	-0.07845	-0.14427	0.49806
927	6	1.047	3.07	-0.00	0.827	0.831	0.824	-0.07721	-0.14273	0.48036
927	7	1.045	4.09	-0.00	0.818	0.824	0.818	-0.08525	-0.14615	0.50264
927	8	1.047	6.15	-0.00	0.801	0.814	0.809	-0.09305	-0.17157	0.52633
927	9	1.057	8.30	-0.00	0.790	0.791	0.791	-0.09743	-0.17704	0.53083
927	10	1.049	10.42	-0.00	0.776	0.779	0.765	-0.10451	-0.18943	0.54986
927	11	1.049	12.42	-0.00	0.772	0.769	0.754	-0.10606	-0.19747	0.54317
928	1	0.999	-4.05	-0.00	0.829	0.993	0.974	0.01512	0.01441	0.14791
928	2	0.999	-3.10	-0.00	1.026	0.993	0.972	0.01389	0.01581	0.15275
928	3	0.999	-2.10	-0.00	1.021	0.991	0.972	0.01120	0.01667	0.16492
928	4	0.997	-1.02	-0.00	1.025	0.993	0.974	0.01227	0.01463	0.16406
928	5	0.997	0.99	-0.00	1.019	0.985	0.974	0.00994	0.01881	0.16804
928	6	0.996	3.02	-0.00	1.020	0.987	0.986	0.01056	0.01842	0.15904
928	7	0.996	4.11	-0.00	1.021	0.980	0.986	0.01110	0.01909	0.15164
928	8	0.995	6.14	-0.00	1.003	0.980	0.978	0.00105	0.01915	0.17394
928	9	0.995	8.32	-0.00	0.977	0.971	0.964	0.00231	0.02292	0.17777
928	10	1.001	10.39	-0.00	0.977	0.950	0.946	0.00131	0.02961	0.19077
928	11	1.001	12.39	-0.00	0.959	0.916	0.916	0.00100	0.04641	0.21291
929	1	0.973	-4.08	-0.00	0.890	0.889	0.885	-0.05917	-0.10847	0.33874

TABLE VB
BALANCE CAVITY AND MODEL BASE PRESSURE COEFFICIENTS AND INCREMENTAL
AXIAL FORCE COEFFICIENTS - THRUST EFFECT PHASE

Run	Pt.	M _c	α	Φ	C _{p,c}	C _{p,b1}	C _{p,b2}	ΔC _{A,c}	ΔC _{A,b}	C _{A,u}
929	2	0.975	-3.04	0.00	0.893	0.891	0.888	-0.057	0.105	0.344
929	3	0.972	-2.08	-0.00	0.893	0.893	0.889	-0.057	0.104	0.338
929	4	0.972	-1.05	-0.00	0.891	0.890	0.886	-0.058	0.106	0.340
929	5	0.971	-1.01	-0.00	0.892	0.890	0.886	-0.058	0.107	0.338
929	6	0.973	3.05	-0.00	0.889	0.887	0.882	-0.060	0.107	0.340
929	7	0.972	3.07	-0.00	0.889	0.887	0.882	-0.060	0.108	0.342
929	8	0.973	4.09	-0.00	0.886	0.885	0.879	-0.062	0.111	0.340
929	10	0.971	6.14	-0.00	0.874	0.872	0.868	-0.064	0.113	0.340
929	11	0.971	8.20	-0.00	0.858	0.857	0.850	-0.067	0.115	0.340
929	12	0.972	10.32	-0.00	0.830	0.825	0.815	-0.076	0.114	0.339
929	13	0.971	12.38	-0.00	0.830	0.825	0.815	-0.076	0.117	0.341
930	3	0.974	-4.08	0.00	1.238	1.184	1.191	0.129	0.161	0.216
930	4	0.973	-3.04	-0.00	1.239	1.183	1.196	0.130	0.161	0.222
930	5	0.974	-2.08	-0.00	1.235	1.193	1.191	0.127	0.166	0.215
930	6	0.974	-1.03	-0.00	1.245	1.193	1.186	0.127	0.166	0.215
930	7	0.974	-0.03	-0.00	1.249	1.180	1.183	0.125	0.166	0.217
930	8	0.972	1.01	-0.00	1.252	1.184	1.184	0.124	0.160	0.220
930	9	0.972	2.07	-0.00	1.257	1.177	1.180	0.124	0.161	0.220
930	10	0.973	3.10	-0.00	1.260	1.174	1.179	0.123	0.167	0.223
930	11	0.975	4.10	-0.00	1.268	1.157	1.145	0.121	0.167	0.223
930	12	0.974	6.29	-0.00	1.263	1.137	1.128	0.114	0.148	0.209
930	13	0.974	10.34	0.00	1.201	1.119	1.082	0.100	0.071	0.165
931	1	0.973	-4.09	0.00	1.143	1.049	0.94	0.077	0.095	0.062
931	2	0.972	-3.04	-0.00	1.155	1.099	0.94	0.077	0.095	0.062
931	3	0.971	-2.07	-0.00	1.147	1.105	0.94	0.079	0.099	0.067
931	4	0.972	-1.04	-0.00	1.146	1.103	0.94	0.079	0.099	0.067
931	5	0.972	-0.02	-0.00	1.148	1.103	0.94	0.080	0.099	0.067
931	6	0.971	1.03	-0.00	1.148	1.101	0.94	0.080	0.099	0.067
931	7	0.974	2.06	-0.00	1.147	1.099	0.94	0.080	0.099	0.067
931	8	0.971	3.09	-0.00	1.150	1.099	0.94	0.081	0.099	0.067
931	9	0.973	4.12	-0.00	1.144	1.097	0.94	0.081	0.099	0.067
931	10	0.975	6.30	-0.00	1.132	1.067	0.94	0.077	0.099	0.067
931	11	0.975	10.35	0.00	1.115	1.069	0.94	0.061	0.099	0.067
931	12	0.975	12.35	0.00	1.115	1.069	0.94	0.061	0.099	0.067
932	1	0.974	-4.09	-0.00	1.046	1.010	0.994	0.025	0.002	0.103

TABLE VB
BALANCE CAVITY AND MODEL BASE PRESSURE COEFFICIENTS AND INCREMENTAL AXIAL FORCE COEFFICIENTS - THRUST EFFECT PHASE

Run	Pt.	M _c	α	φ	C _{p,c}	C _{p,b1}	C _{p,b2}	ΔC _{A,c}	ΔC _{A,b}	C _{A,u}
932	2	0.972	-3.04	-0.00	1.036	1.003	0.982	0.02070	-0.00676	0.10463
932	3	0.972	-2.07	-0.00	1.034	1.004	0.981	0.01900	-0.00688	0.11919
932	4	0.972	-1.02	-0.00	1.034	1.003	0.984	0.01961	-0.00495	0.12190
932	5	0.971	1.00	-0.00	1.032	1.000	0.985	0.01867	-0.00584	0.11876
932	6	0.973	2.04	-0.00	1.033	1.001	0.982	0.01764	-0.00676	0.11963
932	7	0.970	3.06	-0.00	1.034	1.004	0.999	0.01821	-0.00293	0.12016
932	8	0.971	4.07	-0.00	1.028	1.002	0.999	0.01851	0.00099	0.11674
932	9	0.972	6.11	-0.00	1.023	1.002	0.996	0.01304	-0.00045	0.12229
932	10	0.974	8.31	-0.00	1.021	0.997	0.992	0.01355	-0.00489	0.12751
932	11	0.974	10.35	0.00	0.945	0.982	0.875	0.03955	-0.10634	0.14909
932	12	0.946	-4.07	-0.00	1.251	1.179	1.165	0.3289	0.18606	0.286
932	13	0.946	-3.04	-0.00	1.226	1.179	1.176	0.32990	0.18553	0.23905
932	14	0.946	-2.04	-0.00	1.225	1.182	1.176	0.32667	0.18253	0.23549
932	15	0.948	-1.03	-0.00	1.232	1.187	1.179	0.32808	0.18261	0.23748
932	16	0.949	1.01	-0.00	1.236	1.189	1.170	0.33399	0.17992	0.23368
932	17	0.949	3.05	-0.00	1.241	1.174	1.170	0.33599	0.17491	0.23204
932	18	0.949	4.10	-0.00	1.241	1.168	1.163	0.33793	0.16502	0.22647
932	19	0.947	6.14	-0.00	1.247	1.148	1.145	0.34065	0.14800	0.21491
932	20	0.948	8.19	-0.00	1.247	1.129	1.115	0.34176	0.12188	0.20091
932	21	0.948	10.36	0.00	1.190	1.090	1.051	0.31678	0.08741	0.18091
932	22	0.948	12.35	0.00	1.133	1.094	1.051	0.27639	0.07660	0.16264
934	1	0.946	-4.07	-0.00	1.133	1.097	1.085	0.27639	0.09116	0.08138
934	2	0.950	-3.04	-0.00	1.136	1.109	1.086	0.27696	0.09421	0.08050
934	3	0.948	-2.10	-0.00	1.136	1.107	1.089	0.27731	0.09611	0.07702
934	4	0.949	-1.02	-0.00	1.138	1.109	1.086	0.27833	0.09358	0.07554
934	5	0.948	1.00	-0.00	1.137	1.109	1.090	0.27955	0.09401	0.07542
934	6	0.948	3.06	-0.00	1.139	1.109	1.087	0.27867	0.09313	0.07372
934	7	0.948	4.09	-0.00	1.132	1.109	1.087	0.27962	0.09328	0.07360
934	8	0.948	6.13	-0.00	1.132	1.108	1.068	0.27542	0.09238	0.07630
934	9	0.946	8.21	-0.00	1.168	1.085	1.073	0.26669	0.07931	0.06914
934	10	0.950	10.36	0.00	1.103	1.097	0.962	0.25908	0.06896	0.02648
934	11	0.950	12.36	0.00	0.902	0.903	0.897	0.05531	-0.03270	0.01582
935	1	0.950	-4.08	-0.00	0.902	0.903	0.897	-0.05531	-0.10051	0.31039

TABLE VB
BALANCE CAVITY AND MODEL BASE PRESSURE COEFFICIENTS AND INCREMENTAL
AXIAL FORCE COEFFICIENTS - THRUST EFFECT PHASE

Run	Pt.	M _c	α	Φ	C _{p_c}	C _{p_{b1}}	C _{p_{b2}}	ΔC _{A_c}	ΔC _{A_b}	C _{A_u}
935	2	0.948	-3.05	-0.00	0.903	0.903	0.898	0.5530	-0.1008	0.31297
935	3	0.946	-2.105	-0.00	0.903	0.902	0.898	0.5515	-0.0998	0.31143
935	4	0.948	-1.01	-0.00	0.905	0.905	0.900	0.5422	-0.1015	0.31846
935	5	0.949	-0.98	-0.00	0.903	0.902	0.899	0.5473	-0.1008	0.31587
935	6	0.947	3.06	-0.00	0.903	0.902	0.898	0.5512	-0.1012	0.31222
935	7	0.950	4.09	-0.00	0.902	0.901	0.897	0.5504	-0.1017	0.31449
935	8	0.946	6.13	-0.00	0.897	0.891	0.888	0.5543	-0.1017	0.31277
935	9	0.948	8.20	-0.00	0.897	0.893	0.883	0.5468	-0.1102	0.32602
935	10	0.946	10.36	-0.00	0.882	0.877	0.876	0.5448	-0.1253	0.33644
935	11	0.946	12.36	-0.00	0.865	0.856	0.857	0.5770	-0.1449	0.35194
935	12	0.946		-0.00	0.865	0.856	0.857	0.5770	-0.1449	0.35194
936	6	0.899	-4.06	-0.00	1.109	1.075	1.061	0.6934	0.7688	0.5627
936	7	0.898	-3.07	-0.00	1.107	1.076	1.064	0.6993	0.7786	0.5417
936	8	0.897	-2.05	-0.00	1.106	1.074	1.061	0.6891	0.7705	0.5173
936	9	0.897	-1.03	-0.00	1.106	1.072	1.061	0.6925	0.7690	0.4750
936	10	0.896	0.99	-0.00	1.106	1.070	1.062	0.6827	0.7505	0.4627
936	11	0.898	3.06	-0.00	1.106	1.070	1.062	0.6806	0.7533	0.4577
936	12	0.898	4.12	-0.00	1.107	1.070	1.061	0.6867	0.7549	0.4745
936	13	0.897	6.20	-0.00	1.102	1.063	1.051	0.6870	0.7543	0.4709
936	14	0.897	8.24	-0.00	1.090	1.049	1.040	0.6570	0.6322	0.4286
936	15	0.896	10.31	-0.00	1.074	1.007	0.968	0.5768	0.3349	0.2863
936	16	0.897	12.31	-0.00	1.069	0.991	0.950	0.4443	0.0330	0.0463
937	1	0.800	-4.05	-0.00	1.189	1.142	1.149	1.5230	0.2029	0.8233
937	2	0.795	-3.08	-0.00	1.185	1.150	1.156	1.5351	0.2122	0.8452
937	3	0.798	-2.04	-0.00	1.187	1.150	1.161	1.5126	0.2222	0.8949
937	4	0.799	-1.03	-0.00	1.193	1.151	1.161	1.4293	0.2308	0.9427
937	5	0.798	0.99	-0.00	1.196	1.152	1.141	1.5033	0.2143	0.9151
937	6	0.797	3.02	-0.00	1.201	1.153	1.141	1.6078	0.2062	0.8918
937	7	0.798	4.07	-0.00	1.206	1.155	1.139	1.6604	0.1928	0.8601
937	8	0.797	6.10	-0.00	1.208	1.121	1.108	1.6051	0.1722	0.8189
937	9	0.797	8.13	-0.00	1.199	1.116	1.105	1.6949	0.1634	0.7743
937	10	0.797	10.27	-0.00	1.159	1.101	1.041	1.5495	0.1537	0.7138
937	11	0.796	12.27	-0.00	1.159	1.101	1.041	1.5495	0.1537	0.7138
937	12	0.796		-0.00	1.159	1.101	1.041	1.5495	0.1537	0.7138
938	1	0.797	-4.07	-0.00	1.045	1.019	1.008	0.3696	0.2011	0.0513

TABLE VB
BALANCE CAVITY AND MODEL BASE PRESSURE COEFFICIENTS AND INCREMENTAL
AXIAL FORCE COEFFICIENTS - THRUST EFFECT PHASE

Run	Pt.	M_c	α	Φ	C_{Pc}	C_{Pb1}	C_{Pb2}	ΔC_{Ac}	ΔC_{Ab}	C_{Au}
938	3	0.797	-3.03	-0.00	1.048	1.021	1.007	0.03886	0.02105	0.05068
938	4	0.797	-2.05	-0.00	1.046	1.020	1.005	0.03801	0.02140	0.05091
938	5	0.797	-1.02	-0.00	1.043	1.019	1.004	0.03547	0.01674	0.06294
938	6	0.799	0.98	-0.00	1.041	1.017	1.003	0.03307	0.01687	0.06268
938	7	0.798	3.05	-0.00	1.039	1.013	1.004	0.03167	0.01515	0.06760
938	8	0.799	4.05	-0.00	1.036	1.017	1.009	0.02935	0.01607	0.06921
938	10	0.797	6.17	-0.00	1.035	1.017	1.014	0.02861	0.02284	0.06750
938	11	0.799	8.14	-0.00	1.034	1.016	1.019	0.02742	0.02684	0.07100
938	13	0.800	12.28	0.00	1.019	0.980	1.020	0.01544	0.02684	0.09987
939	1	0.394	0.00	-0.00	1.025	1.014	1.013	0.06338	0.06626	0.05466
939	2	0.396	-3.05	-0.00	1.022	1.012	1.013	0.07153	0.07969	0.04537
939	3	0.397	-2.04	-0.00	1.023	1.012	1.012	0.07394	0.07172	0.04547
939	4	0.398	-1.04	-0.00	1.022	1.012	1.012	0.07169	0.07121	0.04077
939	5	0.398	0.97	-0.00	1.021	1.012	1.013	0.06999	0.07030	0.04100
939	6	0.398	3.04	-0.00	1.021	1.012	1.013	0.06857	0.07031	0.04335
939	7	0.399	4.04	-0.00	1.019	1.012	1.014	0.06576	0.07698	0.04367
939	8	0.399	6.12	0.00	1.016	1.011	1.014	0.05122	0.07733	0.04364
939	10	0.398	12.13	0.00	1.016	1.005	1.010	0.05426	0.07156	0.04750
940	3	1.250	4.12	-0.00	1.427	1.359	1.296	0.4019	0.18370	0.3820
940	4	1.249	5.16	-0.00	1.403	1.349	1.295	0.4273	0.19300	0.3898
940	5	1.247	7.13	-0.00	1.392	1.343	1.295	0.4272	0.19729	0.3846
940	6	1.248	10.04	-0.00	1.394	1.343	1.295	0.4295	0.19652	0.3859
940	7	1.249	13.03	-0.00	1.404	1.349	1.295	0.4295	0.19014	0.3896
940	8	1.249	17.06	-0.00	1.417	1.354	1.295	0.4107	0.19969	0.3896
940	10	1.250	25.17	0.00	1.466	1.390	1.290	0.4804	0.15563	0.3329
940	11	1.249	35.46	0.00	1.466	1.390	1.290	0.4804	0.14915	0.3329
940	13	1.249	48.48	0.00	1.465	1.388	1.290	0.4804	0.16342	0.3329
941	16	1.250	-4.06	-0.00	1.115	1.067	1.066	0.03809	0.04512	0.14974

TABLE VB
BALANCE CAVITY AND MODEL BASE PRESSURE COEFFICIENTS AND INCREMENTAL
AXIAL FORCE COEFFICIENTS - THRUST EFFECT PHASE

Run	Pt.	M _c	α	φ	C _{Pc}	C _{Pb1}	C _{Pb2}	ΔC _{Ac}	ΔC _{Ab}	C _{Au}
941	17	1.249	-3.07	-0.00	1.113	1.076	1.060	0.03603	0.04072	0.15097
941	18	1.250	-2.11	-0.00	1.115	1.077	1.063	0.03725	0.04104	0.15297
941	19	1.252	-1.05	-0.00	1.115	1.079	1.065	0.03786	0.04204	0.15284
941	16	1.253	-0.02	-0.00	1.114	1.076	1.059	0.03714	0.04031	0.15103
941	7	1.249	0.98	-0.00	1.114	1.075	1.061	0.03743	0.04110	0.15207
941	8	1.249	0.06	-0.00	1.112	1.074	1.056	0.03686	0.03843	0.15249
941	9	1.250	3.33	-0.00	1.112	1.062	1.057	0.03696	0.03843	0.15216
941	14	1.253	4.16	-0.00	1.102	1.064	1.051	0.03349	0.02907	0.15063
941	15	1.253	6.35	-0.00	1.109	1.064	1.035	0.02991	0.02907	0.15051
941	16	1.251	10.35	-0.00	1.078	1.030	1.017	0.02705	0.01927	0.14185
941			12.46	-0.00				0.02577	0.00963	0.17402
942	1	1.095	-4.09	-0.00	0.830	0.827	0.819	0.07275	0.13443	0.47296
942	2	1.102	-3.05	-0.00	0.833	0.836	0.835	0.06628	0.12185	0.43296
942	3	1.097	-2.09	-0.00	0.836	0.836	0.832	0.06819	0.12552	0.43102
942	4	1.095	-1.03	-0.00	0.842	0.843	0.835	0.06819	0.12552	0.43102
942	5	1.101	0.01	-0.00	0.842	0.843	0.833	0.06661	0.12300	0.44007
942	6	1.095	1.02	-0.00	0.837	0.841	0.824	0.06773	0.12363	0.43977
942	7	1.100	2.12	-0.00	0.837	0.841	0.824	0.06621	0.12363	0.43977
942	8	1.099	3.15	-0.00	0.803	0.806	0.819	0.07355	0.15726	0.45551
942	9	1.103	4.32	-0.00	0.757	0.761	0.769	0.08355	0.17182	0.47814
942	10	1.101	6.40	-0.00	0.757	0.761	0.760	0.09280	0.18994	0.50472
942	11	1.101	10.42	-0.00	0.744	0.756	0.739	0.10896	0.18994	0.52301
943	1	1.098	-4.08	-0.00	1.168	1.145	1.143	0.08026	0.10257	0.00030
943	2	1.109	-3.06	-0.00	1.190	1.151	1.143	0.08011	0.10906	0.00150
943	3	1.100	-1.06	-0.00	1.190	1.146	1.133	0.08075	0.10648	0.00473
943	4	1.098	0.01	-0.00	1.195	1.149	1.136	0.08407	0.10543	0.00508
943	5	1.098	1.05	-0.00	1.200	1.152	1.136	0.08409	0.10542	0.00504
943	6	1.102	2.08	-0.00	1.206	1.143	1.136	0.08733	0.10634	0.00620
943	7	1.102	3.16	-0.00	1.210	1.134	1.138	0.08727	0.10591	0.00705
943	8	1.102	4.60	-0.00	1.217	1.128	1.135	0.08552	0.09979	0.00562
943	9	1.101	6.31	-0.00	1.193	1.126	1.137	0.08597	0.09205	0.00114
943	10	1.105	10.43	-0.00	1.183	1.111	1.115	0.07715	0.07798	0.00114
944	1	1.101	-4.09	0.00	1.215	1.154	1.149	0.09146	0.11461	0.03023

TABLE VB
 BALANCE CAVITY AND MODEL BASE PRESSURE COEFFICIENTS AND INCREMENTAL
 AXIAL FORCE COEFFICIENTS - THRUST EFFECT PHASE

Run	Pt.	M _c	α	φ	C _{p c}	C _{p b1}	C _{p b2}	ΔC _{A c}	ΔC _{A b}	C _{A u}
944	3	1.098	-5.04	0.00	1.209	1.156	1.163	0.08945	0.12208	0.02294
944	4	1.099	-2.04	0.00	1.206	1.164	1.159	0.08768	0.12223	0.02288
944	5	1.103	-1.01	0.00	1.213	1.165	1.152	0.08878	0.12008	0.02071
944	6	1.098	1.02	0.00	1.219	1.166	1.155	0.09083	0.12201	0.02293
944	7	1.102	2.04	0.00	1.224	1.156	1.157	0.09308	0.11288	0.02753
944	8	1.102	3.10	0.00	1.227	1.142	1.156	0.09477	0.11288	0.02756
944	9	1.102	4.16	0.00	1.229	1.138	1.142	0.09622	0.10613	0.02746
944	10	1.104	6.26	0.00	1.222	1.135	1.122	0.09707	0.10674	0.02749
944	11	1.104	8.33	0.00	1.220	1.134	1.148	0.09397	0.10431	0.02511
944	12	1.100	12.43	0.00	1.212	1.130	1.143	0.09010	0.10916	0.02746
945	3	1.095	4.07	0.00	1.364	1.241	1.236	0.15472	0.16032	0.18738
945	4	1.095	5.06	0.00	1.356	1.260	1.266	0.15174	0.19927	0.19562
945	5	1.097	6.07	0.00	1.352	1.284	1.289	0.15074	0.21065	0.20302
945	6	1.096	7.01	0.00	1.349	1.302	1.302	0.14906	0.22426	0.20621
945	7	1.099	8.04	0.00	1.364	1.286	1.299	0.15478	0.22178	0.20621
945	8	1.097	9.09	0.00	1.379	1.260	1.267	0.16946	0.20398	0.20291
945	9	1.095	10.13	0.00	1.400	1.240	1.247	0.17243	0.18516	0.19086
945	10	1.095	11.16	0.00	1.420	1.206	1.219	0.17911	0.16177	0.19086
945	11	1.095	12.32	0.00	1.406	1.241	1.248	0.17284	0.16193	0.20422
945	12	1.098	12.32	0.01	1.355	1.221	1.169	0.15143	0.18824	0.22773
946	4	1.000	4.07	0.00	1.053	1.029	1.006	0.02764	0.01663	0.08064
946	5	1.000	5.11	0.00	1.043	1.004	0.991	0.02295	0.01346	0.07343
946	6	1.000	6.13	0.00	1.038	1.009	0.990	0.02259	0.00907	0.06672
946	7	1.000	7.15	0.00	1.033	1.003	0.982	0.01709	0.00540	0.05921
946	8	1.000	8.16	0.00	1.028	1.000	0.980	0.01449	0.00406	0.05412
946	9	1.000	9.19	0.00	1.027	1.003	0.981	0.01357	0.00422	0.05054
946	10	1.003	10.21	0.00	1.030	1.007	0.987	0.01546	0.00804	0.05380
946	11	1.000	11.21	0.00	1.019	1.009	0.987	0.01546	0.00804	0.05380
946	12	1.000	12.31	0.00	1.019	1.009	0.970	0.00436	0.00957	0.05144
946	13	1.000	12.37	0.00	1.011	1.009	0.960	0.00418	0.02224	0.05469
947	1	0.898	-4.04	0.00	1.366	1.218	1.281	0.23320	0.26283	0.48533

TABLE VB
BALANCE CAVITY AND MODEL BASE PRESSURE COEFFICIENTS AND INCREMENTAL AXIAL FORCE COEFFICIENTS - THRUST EFFECT PHASE

Run	Pt.	M _c	α	φ	C _p c	C _p b ₁	C _p b ₂	ΔC _A c	ΔC _A b	C _A u
947	2	0.899	-3.02	0.00	1.378	1.250	1.275	0.24094	0.29735	-0.50227
947	3	0.900	-2.05	0.00	1.373	1.269	1.267	0.23692	0.32598	-0.51126
947	4	0.899	-1.05	0.00	1.354	1.313	1.317	0.22192	0.35965	-0.50673
947	5	0.897	-0.05	-0.00	1.374	1.293	1.315	0.23696	0.34282	-0.51595
947	6	0.901	0.97	-0.00	1.402	1.245	1.270	0.25651	0.34201	-0.51274
947	7	0.899	3.05	-0.00	1.444	1.217	1.239	0.26863	0.25783	-0.50935
947	8	0.898	4.06	-0.00	1.441	1.215	1.211	0.28085	0.25817	-0.51832
947	9	0.898	6.11	0.00	1.421	1.256	1.267	0.26890	0.26476	-0.52916
947	10	0.898	10.16	0.00	1.421	1.256	1.267	0.26890	0.31304	-0.52516
947	11	0.899	12.33	0.00	1.386	1.301	1.272	0.23266	0.32091	-0.52831
947	12	0.896	12.33	0.00	1.386	1.301	1.272	0.23266	0.32091	-0.52831
948	1	0.899	0.05	0.00	1.173	1.320	1.27	0.11052	0.4654	-0.18316
948	2	0.895	-3.09	-0.00	1.173	1.143	1.136	0.11052	0.15694	-0.18459
948	3	0.899	-2.06	-0.00	1.178	1.143	1.133	0.11323	0.15564	-0.18531
948	4	0.899	-1.06	-0.00	1.182	1.143	1.129	0.11583	0.15480	-0.18599
948	5	0.899	0.97	-0.00	1.185	1.133	1.130	0.11470	0.15193	-0.18026
948	6	0.898	3.06	-0.00	1.185	1.133	1.130	0.11470	0.15193	-0.17707
948	7	0.898	4.06	-0.00	1.180	1.129	1.119	0.12165	0.14952	-0.17492
948	8	0.899	6.10	0.00	1.192	1.125	1.097	0.12165	0.14952	-0.16935
948	9	0.898	10.32	0.00	1.161	1.131	1.086	0.10932	0.11853	-0.16354
948	10	0.898	12.32	0.00	1.146	1.119	1.069	0.10932	0.11853	-0.16354
948	11	0.898	12.32	0.00	1.146	1.119	1.069	0.10932	0.11853	-0.16354
948	12	0.898	12.32	0.00	1.146	1.119	1.069	0.10932	0.11853	-0.16354
949	14	0.900	1.17	0.00	1.073	0.954	0.937	0.04722	0.05188	0.00207
949	15	0.898	-3.09	-0.00	1.074	1.046	1.035	0.04655	0.04571	0.01563
949	16	0.898	-2.09	-0.00	1.072	1.046	1.035	0.04655	0.04571	0.01273
949	17	0.898	-1.04	-0.00	1.071	1.045	1.034	0.04556	0.04166	0.02222
949	18	0.898	0.98	-0.00	1.068	1.042	1.030	0.04576	0.04185	0.02246
949	19	0.900	3.05	0.00	1.074	1.042	1.030	0.04701	0.04897	0.02091
949	20	0.898	4.07	0.00	1.069	1.042	1.030	0.04701	0.04897	0.02091
949	21	0.898	6.10	0.00	1.059	1.030	1.013	0.03878	0.03445	0.02077
949	22	0.898	10.34	0.00	1.056	1.030	1.013	0.03529	0.03459	0.02942
949	23	0.897	12.34	0.00	1.051	1.031	1.006	0.03529	0.03459	0.02942
949	24	0.899	12.34	0.00	1.051	1.031	1.006	0.03529	0.03459	0.02942
949	25	0.899	12.34	0.00	1.051	1.031	1.006	0.03529	0.03459	0.02942
949	26	0.899	12.34	0.00	1.051	1.031	1.006	0.03529	0.03459	0.02942
950	1	0.899	-4.07	-0.00	0.919	0.919	0.912	-0.05096	-0.09517	0.28594

TABLE VB
BALANCE CAVITY AND MODEL BASE PRESSURE COEFFICIENTS AND INCREMENTAL
AXIAL FORCE COEFFICIENTS - THRUST EFFECT PHASE

Run	Pt.	M _c	α	φ	C _{p c}	C _{p b 1}	C _{p b 2}	ΔC _{A c}	ΔC _{A b}	C _{A u}
950	2	0.899	-3.03	-0.00	0.922	0.920	0.915	-0.049	0.927	0.288
950	3	0.899	-2.05	-0.00	0.923	0.924	0.917	-0.049	0.900	0.284
950	4	0.901	-1.01	-0.00	0.923	0.924	0.916	-0.049	0.891	0.287
950	5	0.898	-0.99	-0.00	0.923	0.925	0.917	-0.049	0.891	0.288
950	6	0.899	3.06	-0.00	0.922	0.923	0.917	-0.049	0.896	0.293
950	7	0.900	4.06	-0.00	0.919	0.921	0.914	-0.050	0.909	0.292
950	8	0.900	6.10	-0.00	0.904	0.901	0.900	-0.061	0.911	0.300
950	10	0.896	8.24	-0.00	0.902	0.901	0.899	-0.061	0.909	0.300
950	11	0.899	12.33	-0.00	0.902	0.906	0.882	-0.061	0.918	0.283
951	4	0.800	4.06	0.00	1.273	1.53	1.73	0.219	3.37	1.17
951	5	0.798	-3.09	0.00	1.264	1.54	1.72	0.215	3.77	1.19
951	6	0.798	-2.07	0.00	1.254	1.20	1.20	0.205	4.80	1.33
951	7	0.797	-1.05	0.00	1.245	1.22	1.22	0.198	4.91	1.34
951	8	0.795	-0.97	-0.00	1.255	1.20	1.21	0.200	4.67	1.35
951	10	0.796	3.03	0.00	1.272	1.86	1.70	0.233	4.68	1.35
951	11	0.794	4.07	0.00	1.304	1.63	1.46	0.237	4.60	1.37
951	13	0.796	6.16	0.00	1.306	1.50	1.46	0.230	4.60	1.37
951	14	0.796	8.21	0.00	1.254	1.20	1.17	0.217	4.60	1.37
951	16	0.796	12.27	0.00	1.254	1.20	1.17	0.217	4.60	1.37
952	17	0.797	4.07	-0.00	1.110	0.87	0.80	0.091	1.24	0.94
952	19	0.799	-3.05	-0.00	1.119	0.92	0.86	0.095	1.25	1.03
952	20	0.797	-2.08	-0.00	1.122	0.91	0.86	0.098	1.25	1.05
952	21	0.796	-1.04	-0.00	1.122	0.80	0.79	0.087	1.21	0.93
952	22	0.799	3.05	-0.00	1.122	0.81	0.76	0.091	1.22	1.02
952	23	0.798	4.06	-0.00	1.120	0.72	0.76	0.097	1.15	0.92
952	24	0.796	6.08	0.00	1.121	0.59	0.58	0.090	1.05	0.80
952	26	0.798	8.13	0.00	1.119	0.76	0.58	0.096	1.02	0.81
952	29	0.798	12.26	0.00	1.104	0.65	0.46	0.084	0.92	0.71
953	1	0.796	-4.04	-0.00	1.042	1.02	1.01	0.034	0.225	0.052

TABLE VB
BALANCE CAVITY AND MODEL BASE PRESSURE COEFFICIENTS AND INCREMENTAL
AXIAL FORCE COEFFICIENTS - THRUST EFFECT PHASE

Run	Pt.	M _c	α	φ	C _{p c}	C _{p b1}	C _{p b2}	ΔC _{A c}	ΔC _{A b}	C _{A u}
953	3	0.795	-3.06	-0.00	1.043	1.020	1.007	0.3559	0.1604	0.6009
953	4	0.798	-2.03	-0.00	1.046	1.022	1.011	0.3775	0.2264	0.5593
953	5	0.798	-1.06	-0.00	1.040	1.021	1.013	0.3287	0.0181	0.6856
953	6	0.798	0.03	-0.00	1.037	1.019	1.010	0.3057	0.0211	0.6953
953	7	0.797	3.04	-0.00	1.036	1.020	1.017	0.3046	0.0278	0.7020
953	8	0.795	4.07	-0.00	1.039	1.019	1.013	0.2945	0.0278	0.6304
953	9	0.798	6.10	-0.00	1.035	1.014	0.995	0.2727	0.0154	0.6807
953	10	0.794	10.21	0.00	1.027	1.011	0.985	0.2246	0.0024	0.6487
953	11	0.793	12.29	0.00	1.028	1.017	0.984	0.2361	0.0011	0.5389
954	1	0.796	0.07	-0.00	0.940	0.937	0.935	0.4817	0.9328	0.2866
954	2	0.795	-3.07	-0.00	0.941	0.938	0.937	0.4773	0.9146	0.2866
954	3	0.796	-2.06	-0.00	0.943	0.941	0.937	0.4510	0.8670	0.2904
954	4	0.795	-1.01	-0.00	0.943	0.941	0.937	0.4526	0.8673	0.2874
954	5	0.794	0.00	-0.00	0.941	0.941	0.936	0.4531	0.8680	0.2914
954	6	0.794	3.05	-0.00	0.941	0.939	0.933	0.4513	0.8689	0.2913
954	7	0.793	4.08	-0.00	0.934	0.936	0.930	0.4593	0.9211	0.2933
954	8	0.793	6.16	-0.00	0.931	0.927	0.924	0.5578	1.0715	0.2950
954	9	0.793	10.29	0.00	0.930	0.928	0.916	0.5651	1.1186	0.2864
955	1	0.392	-4.03	0.00	1.039	1.028	1.029	1.3259	1.7155	0.1918
955	2	0.392	-3.04	0.00	1.039	1.030	1.030	1.3231	1.7966	0.1921
955	3	0.392	-2.05	0.00	1.040	1.031	1.030	1.3226	1.7930	0.2098
955	4	0.392	-1.02	0.00	1.041	1.031	1.030	1.3348	1.8299	0.1866
955	5	0.392	0.02	0.00	1.041	1.030	1.030	1.3360	1.8064	0.1853
955	6	0.392	3.02	0.00	1.042	1.027	1.027	1.4201	1.7669	0.1914
955	7	0.392	4.06	0.00	1.042	1.023	1.021	1.5969	1.6397	0.1753
955	8	0.392	6.10	0.00	1.042	1.023	1.021	1.4603	1.5350	0.1753
955	9	0.392	10.13	0.00	1.042	1.024	1.015	1.4617	1.3350	0.1827
955	10	0.392	12.13	0.00	1.039	1.020	1.013	1.4064	1.2864	0.1708
956	1	0.392	-4.01	0.00	0.998	0.990	0.989	0.0038	-0.0618	0.2316

TABLE VB
 BALANCE CAVITY AND MODEL BASE PRESSURE COEFFICIENTS AND INCREMENTAL
 AXIAL FORCE COEFFICIENTS - THRUST EFFECT PHASE

Run	Pt.	M _c	α	Φ	C _{p_c}	C _{p_{b1}}	C _{p_{b2}}	ΔC _{A_c}	ΔC _{A_b}	C _{A_u}	
956	3	0.393	-3.03	0.00	0.999	0.990	0.989	0.000	0.5898	0.000	3.120
956	4	0.394	-2.10	0.00	0.998	0.989	0.988	0.004	0.5522	0.000	3.239
956	5	0.395	-1.04	0.00	0.997	0.992	0.992	0.007	0.5133	0.000	3.350
956	6	0.393	0.00	0.00	0.997	0.990	0.991	0.008	0.4625	0.000	3.418
956	7	0.394	1.99	0.00	0.997	0.990	0.991	0.009	0.4365	0.000	3.520
956	8	0.395	4.01	0.00	0.996	0.991	0.991	0.010	0.5107	0.000	3.635
956	9	0.395	6.07	0.00	0.995	0.990	0.990	0.013	0.5182	0.000	3.752
956	10	0.394	8.11	0.00	0.993	0.989	0.988	0.021	0.5664	0.000	3.874
956	11	0.393	10.12	0.00	0.995	0.991	0.990	0.015	0.5219	0.000	3.992
956	12	0.393	12.12	0.00	0.988	0.985	0.986	0.038	0.6389	0.000	4.116
957	3	0.391	-4.03	0.00	0.988	0.985	0.986	0.037	0.6379	0.000	4.240
957	4	0.392	-3.07	0.00	0.988	0.985	0.986	0.033	0.6170	0.000	4.364
957	5	0.392	-2.04	0.00	0.989	0.986	0.987	0.032	0.7853	0.000	4.488
957	6	0.392	-1.05	0.00	0.988	0.986	0.986	0.030	0.8037	0.000	4.612
957	7	0.392	0.98	0.00	0.988	0.986	0.986	0.037	0.8033	0.000	4.736
957	8	0.393	3.95	0.00	0.988	0.985	0.985	0.043	0.8456	0.000	4.860
957	9	0.393	5.97	0.00	0.987	0.984	0.985	0.041	0.8530	0.000	4.984
957	10	0.393	8.01	0.00	0.986	0.983	0.984	0.043	0.9904	0.000	5.108
957	11	0.392	10.11	0.00	0.986	0.983	0.983	0.044	0.9907	0.000	5.232
957	12	0.392	12.11	0.00	0.985	0.982	0.981	0.049	1.0819	0.000	5.356

KEY TO TABLE VI
 CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	M_c	α	Φ	δ_{c1}	δ_{c2}	δ_{c3}	δ_{c4}		
7										
8		$C_{N_{c1}}$	$C_{HM_{c1}}$	$C_{BM_{c1}}$	CPX_{c1}	$C_{N_{c2}}$	$C_{HM_{c2}}$	$C_{BM_{c2}}$	CPX_{c2}	CPY_{c2}
9		$C_{N_{c3}}$	$C_{HM_{c3}}$	$C_{BM_{c3}}$	CPX_{c3}	$C_{N_{c4}}$	$C_{HM_{c4}}$	$C_{BM_{c4}}$	CPX_{c4}	CPY_{c4}

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																	
20	4	0.899	-3.08	-0.003	-0.1841	0.04	-0.04	0.03	-0.02	0.0642	0.6823	0.0030	-0.0001	0.0003	-0.0295	-0.0003	-0.0040	0.1154	0.6753
		-0.0012	0.0000	-0.0002	-0.6127	1.0561	-0.0299	-0.0006	-0.0040										
20	5	0.901	-1.04	-0.004	-0.1053	0.05	-0.01	0.02	-0.00	0.0519	0.6989	0.0030	-0.0001	0.0004	-0.0073	-0.0010	0.0010	0.3616	0.7600
		-0.0011	0.0000	-0.0003	-0.7675	1.3901	-0.0061	-0.0004	-0.0009										
20	6	0.900	-0.04	-0.004	-0.0491	0.05	-0.00	0.01	0.00	5.9491	0.1749	0.0036	-0.0000	0.0004	0.0002	0.0003	0.0003	-0.4216	0.5390
		-0.0009	0.0002	-0.0002	-1.1013	1.3550	0.0028	-0.0002	0.0000										
20	7	0.901	0.52	-0.004	-0.0510	0.05	-0.00	0.02	0.00	0.6223	0.7244	0.0036	-0.0001	0.0004	0.0040	0.0005	0.0007	-0.0076	0.6917
		-0.0012	0.0000	-0.0002	-0.7369	1.1416	0.0054	-0.0000	0.0000										
20	8	0.900	1.01	-0.005	0.0776	0.05	0.01	0.01	0.01	0.3262	0.6684	0.0035	0.0000	0.0005	0.0094	0.0006	0.0012	0.0780	0.6543
		-0.0011	0.0002	-0.0002	-1.1665	0.9678	0.0090	0.0001	0.0001										
20	9	0.899	3.13	-0.003	0.0157	0.05	0.03	0.01	0.02	0.1274	0.6712	0.0031	0.0000	0.0003	0.0334	0.0008	0.0044	0.0403	0.6730
		-0.0010	0.0002	-0.0002	-1.1882	1.2096	0.0334	0.0002	0.0002										
20	10	0.900	6.22	-0.002	-0.2113	0.04	0.07	0.03	0.05	0.1178	0.6184	0.0016	-0.0000	0.0002	0.0590	0.0013	0.0073	0.0726	0.6110
		-0.0011	0.0003	-0.0001	-1.3867	0.7069	0.0609	0.0008	0.0074										
20	11	0.897	9.33	-0.001	-0.4403	0.04	0.08	0.01	0.07	0.0768	0.5869	0.0014	-0.0001	0.0001	0.0799	0.0012	0.0093	0.0506	0.5780
		-0.0012	0.0002	-0.0001	-1.1858	0.6173	0.0802	0.0008	0.0092										
20	12	0.899	12.47	-0.001	-0.3932	0.04	0.08	0.02	0.07	0.0345	0.5740	0.0016	-0.0001	0.0001	0.0902	0.0006	0.0103	0.0111	0.5703
		-0.0020	0.0002	-0.0002	-0.5913	0.7091	0.0903	0.0002	0.0103										
20	13	0.900	-0.02	-0.004	-0.0292	0.05	-0.00	0.02	-0.00	3.1327	-0.7131	0.0036	-0.0000	0.0004	0.0004	0.0002	-0.0001	-0.5443	0.3413
		-0.0014	0.0003	-0.0001	-1.1798	0.6248	0.0025	-0.0002	0.0001										
21	3	0.899	-3.10	-0.004	-0.1065	0.05	-3.13	0.01	-3.00	0.0733	0.6513	0.0031	-0.0000	0.0004	0.0485	0.0007	-0.0063	0.1238	0.6554
		-0.0011	0.0001	-0.0002	-0.7721	1.1793	-0.0458	-0.0011	-0.0058										

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key													
21	0	0.400	-0.0031	-1.000	-0.0001	-0.0004	-0.0944	0.7231	-0.0278	-3.10	0.01	-2.99	0.0573	0.6846	
	6	0.0031	0.0000	0.0000	0.0000	0.0004	-0.0765	1.0568	-0.0263	-0.0003	-0.0006	-0.0038	0.1249	0.6810	
	9	-0.0011	0.0001	0.0000	-0.0002	-0.0002	-0.0701	0.6661	-0.0159	-3.09	0.01	-2.98	0.0686	0.6975	
21	5	0.899	-0.0034	-0.0002	-0.0002	-0.0004	-0.6867	0.8303	-0.0145	-0.0005	-0.0022	0.1946	0.7167		
	6	0.0034	0.0000	0.0000	0.0002	0.0004	-0.0731	0.6871	-0.0287	-3.10	0.01	-2.99	0.0613	0.6951	
	9	-0.0015	0.0002	0.0000	-0.0002	-0.0005	-0.9025	0.8844	-0.0278	-0.0006	-0.0038	0.1216	0.6940		
21	6	0.899	-0.0036	-0.0002	-0.0002	-0.0004	-0.0362	0.6873	-0.0101	-3.08	0.01	-2.98	0.0980	0.7235	
	7	0.0036	0.0000	0.0000	0.0002	0.0004	-1.0852	0.7828	-0.0099	-0.0004	-0.0014	0.2444	0.7551		
	9	-0.0014	0.0002	0.0000	-0.0002	-0.0005	-0.0319	0.6618	-0.0064	-3.08	0.01	-2.96	0.0254	0.7244	
21	7	0.901	-0.0035	-0.0002	-0.0002	-0.0004	-1.0454	0.9250	-0.0051	-0.0004	-0.0009	0.3902	0.8755		
	8	0.0035	0.0000	0.0000	0.0002	0.0004	-0.0153	0.6005	-0.0113	0.01	-2.94	0.2948	0.6771		
	9	-0.0012	0.0002	0.0000	-0.0002	-0.0005	-1.1978	0.6974	0.0113	0.0002	0.0014	0.1203	0.6210		
21	8	0.898	-0.0032	-0.0002	-0.0002	-0.0004	-0.0672	0.7318	-0.0419	-3.01	0.02	-2.92	0.1358	0.6532	
	9	0.0032	0.0000	0.0000	0.0002	0.0004	-4.8533	0.7569	0.0471	0.0005	0.0062	0.0540	0.6574		
	9	-0.0011	0.0002	0.0000	-0.0002	-0.0005	-0.3862	0.2331	0.0691	0.0008	0.0081	0.1014	0.5905		
21	9	0.898	-0.0021	-0.0003	-0.0003	-0.0004	-1.6618	0.6933	-0.0674	-2.98	0.02	-2.89	0.0702	0.5701	
	7	0.0021	0.0000	0.0000	0.0003	0.0004	-0.3041	0.4709	0.0857	0.0012	0.0097	0.0399	0.5706		
	9	-0.0003	0.0003	0.0000	-0.0003	-0.0004	-0.0386	0.6864	-0.0162	-0.0006	-0.0024	0.1893	0.7321		
21	10	0.898	-0.0021	-0.0003	-0.0003	-0.0004	-1.1054	0.6864	-0.0162	-0.0006	-0.0024	0.1893	0.7321		
	7	0.0021	0.0000	0.0000	0.0003	0.0004	-0.1110	0.7977	-0.0376	-0.0003	-0.0051	0.0490	0.6793		
	9	-0.0003	0.0003	0.0000	-0.0003	-0.0004	-1.1276	2.1978	-0.0372	-0.0006	-0.0049	0.0919	0.6671		
21	11	0.899	-0.0017	-0.0003	-0.0003	-0.0004	-0.0386	0.6933	-0.0674	-2.97	0.02	-2.90	0.0702	0.5706	
	7	0.0017	0.0000	0.0000	0.0003	0.0004	-0.3041	0.4709	0.0857	0.0012	0.0097	0.0399	0.5706		
	9	-0.0010	0.0003	0.0000	-0.0003	-0.0004	-0.0386	0.6864	-0.0162	-0.0006	-0.0024	0.1893	0.7321		
21	12	0.898	-0.0019	-0.0003	-0.0003	-0.0004	-1.1054	0.6864	-0.0162	-0.0006	-0.0024	0.1893	0.7321		
	7	0.0019	0.0000	0.0000	0.0003	0.0004	-0.1110	0.7977	-0.0376	-0.0003	-0.0051	0.0490	0.6793		
	9	-0.0014	0.0003	0.0000	-0.0003	-0.0004	-1.1276	2.1978	-0.0372	-0.0006	-0.0049	0.0919	0.6671		
21	13	0.899	-0.0038	-0.0003	-0.0003	-0.0004	-0.0386	0.6933	-0.0674	-2.97	0.02	-2.90	0.0702	0.5706	
	7	0.0038	0.0000	0.0000	0.0003	0.0004	-0.1110	0.7977	-0.0376	-0.0003	-0.0051	0.0490	0.6793		
	9	-0.0015	0.0003	0.0000	-0.0003	-0.0004	-1.1276	2.1978	-0.0372	-0.0006	-0.0049	0.0919	0.6671		
22	4	0.900	-0.0028	-0.0001	-0.0001	-0.0002	-0.0386	0.6933	-0.0674	-2.97	0.02	-2.90	0.0702	0.5706	
	7	0.0028	0.0000	0.0000	0.0001	0.0002	-0.1110	0.7977	-0.0376	-0.0003	-0.0051	0.0490	0.6793		
	9	-0.0006	0.0001	0.0000	-0.0001	-0.0002	-1.1276	2.1978	-0.0372	-0.0006	-0.0049	0.0919	0.6671		

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																						
22	5	7	0.898	-1.06	-0.00	0.05	-1.05	0.01	-1.04	0.0613	0.6587	0.0035	0.0000	0.0004	-0.0526	0.6823	-0.0144	-0.0001	-0.0004	-0.0001	-0.0005	-0.0019	0.0613	0.6862	
		8	-0.0003	0.0001	0.0002	3.0641	-0.0144	-0.0005	-0.0019	0.1873															
		9																							
22	6	7	0.897	-0.02	-0.00	0.05	-1.04	0.01	-1.02	-0.1468	0.6313	0.0030	0.0000	0.0004	-0.0326	0.7615	-0.0045	-0.0001	-0.0004	-0.0001	-0.0005	-0.0001	-0.1468	0.6313	
		8	-0.0007	0.0001	0.0002	1.6358	-0.0007	-0.0004	-0.0001	2.7430															
		9																							
22	7	7	0.898	0.52	-0.00	0.05	-1.03	0.01	-1.02	-1.6937	0.5395	0.0035	0.0000	0.0004	-0.0208	0.6657	-0.0009	-0.0001	-0.0004	-0.0001	-0.0005	-0.0001	-1.6937	0.5395	
		8	-0.0004	0.0001	0.0002	2.7559	0.0013	-0.0002	0.0001	1.0530															
		9																							
22	8	7	0.898	1.00	-0.00	0.05	-1.03	0.01	-1.01	1.1386	0.6697	0.0032	0.0000	0.0004	-0.0247	0.6689	0.0023	0.0005	-0.0005	-0.0005	-0.0005	0.0004	1.1386	0.6697	
		8	-0.0007	0.0001	0.0002	1.4802	0.0045	-0.0000	0.0005	-0.0898															
		9																							
22	9	7	0.898	3.11	-0.00	0.05	-1.00	0.01	-1.00	0.1663	0.6814	0.0023	0.0000	0.0003	0.0283	0.8240	0.0264	0.0008	-0.0002	-0.0002	-0.0002	0.0036	0.1663	0.6814	
		8	-0.0007	0.0001	0.0002	1.5102	0.0268	0.0002	0.0035	0.0464															
		9																							
22	10	7	0.898	6.19	-0.00	0.04	-0.97	0.01	-0.97	0.1336	0.6271	0.0017	0.0000	0.0002	-0.2911	0.7005	0.0527	0.0014	-0.0008	-0.0008	-0.0009	0.0066	0.1336	0.6271	
		8	-0.0006	0.0002	0.0001	1.1843	0.0557	0.0000	0.0069	0.0738															
		9																							
22	11	7	0.898	9.29	-0.00	0.04	-0.95	0.01	-0.95	0.0838	0.5969	0.0018	0.0000	0.0003	-0.1952	0.8741	0.0762	0.0012	-0.0009	-0.0009	-0.0009	0.0091	0.0838	0.5969	
		8	-0.0008	0.0002	0.0001	0.7020	0.0763	0.0007	0.0089	0.0514															
		9																							
22	12	7	0.898	12.46	-0.00	0.05	-0.95	0.01	-0.95	0.0452	0.5793	0.0013	0.0000	0.0002	-0.4482	0.9263	0.0688	0.0008	-0.0002	-0.0002	-0.0002	0.0103	0.0452	0.5793	
		8	-0.0011	0.0002	0.0001	0.7353	0.0901	0.0002	0.0103	0.0153															
		9																							
22	13	7	0.900	-0.02	-0.00	0.05	-1.05	0.01	-1.03	-0.0811	0.7345	0.0034	0.0000	0.0004	-0.0135	0.6961	-0.0044	0.0000	-0.0004	-0.0005	-0.0005	0.0006	-0.0811	0.7345	
		8	-0.0007	0.0002	0.0001	1.3032	-0.0030	-0.0004	-0.0005	0.7467															
		9																							
23	1	7	0.899	-3.09	-0.00	0.04	1.01	0.01	1.01	0.0600	0.6829	0.0028	0.0000	0.0004	-0.1687	0.7178	-0.0223	0.0002	-0.0002	-0.0002	-0.0003	0.0030	0.0600	0.6829	
		8	-0.0014	0.0002	0.0002	0.9467	-0.0221	-0.0006	-0.0030	0.1417															
		9																							
23	2	7	0.899	-1.05	-0.00	0.05	1.03	0.01	1.03	-0.3160	0.6308	0.0031	0.0000	0.0004	-0.0644	0.7343	-0.0029	0.0001	-0.0003	-0.0003	-0.0003	0.0000	-0.3160	0.6308	
		8	-0.0010	0.0002	0.0002	1.1195	0.0003	-0.0003	-0.0003	-5.8482															
		9																							

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																		
23	5	0.898	-0.01	-0.00	0.05	1.05	0.01	1.04	0.5089	0.7250	0.0034	-0.0000	0.0004	-0.0135	0.6961	0.0056	0.0005	0.0008	0.0151	0.6449
	8	-0.0009	-0.0002	-0.0001	1.0195	0.0072	0.0000	0.0009	0.0000											
	9																			
23	4	0.899	0.53	0.00	0.04	1.05	0.01	1.04	0.2706	0.7107	0.0033	-0.0000	0.0004	-0.0260	0.6528	0.0113	0.0006	0.0016	0.0810	0.6556
	6	-0.0010	-0.0002	-0.0002	1.1137	0.0121	0.0001	0.0015	0.0000											
	9																			
23	5	0.898	1.04	0.00	0.05	1.06	0.01	1.05	0.1968	0.6836	0.0032	0.0000	0.0004	0.0159	0.6663	0.0175	0.0006	0.0023	0.0536	0.6689
	8	-0.0009	0.0002	-0.0002	1.1354	0.0172	0.0001	0.0023	0.0000											
	9																			
23	6	0.896	3.13	0.00	0.05	1.09	0.01	1.06	0.1279	0.6703	0.0023	0.0000	0.0003	0.0457	0.8127	0.0401	0.0010	0.0053	0.0425	0.6710
	8	-0.0011	0.0002	-0.0002	1.0937	0.0405	0.0003	0.0054	0.0000											
	9																			
23	7	0.898	6.22	0.00	0.04	1.12	0.01	1.09	0.1075	0.6144	0.0020	0.0000	0.0002	-0.0981	0.7064	0.0648	0.0013	0.0079	0.0783	0.6030
	8	-0.0008	-0.0002	-0.0001	1.0472	0.0646	0.0010	0.0077	0.0000											
	9																			
23	8	0.898	9.32	0.00	0.04	1.13	0.02	1.10	0.0775	0.5840	0.0017	0.0000	0.0002	-0.2733	0.7848	0.832	0.0012	0.0094	0.0514	0.5765
	8	-0.0007	0.0003	-0.0001	0.7261	0.0815	0.0008	0.0094	0.0000											
	9																			
23	9	0.898	12.47	0.00	0.04	1.12	0.02	1.09	0.0269	0.5774	0.0016	0.0000	0.0002	-0.3582	0.7448	0.908	0.0004	0.0104	0.0039	0.5731
	8	-0.0004	-0.0002	-0.0000	1.0190	0.0913	0.0000	0.0104	0.0000											
	9																			
23	10	0.897	-0.02	0.00	0.04	1.05	0.01	1.04	0.4844	0.5981	0.0032	0.0000	0.0004	-0.0305	0.6678	0.0055	0.0005	0.0009	0.0072	0.5981
	8	-0.0011	0.0003	-0.0001	0.8067	0.0076	0.0000	0.0009	0.0000											
	9																			
24	1	0.897	-3.07	0.09	0.04	3.16	0.03	3.15	0.1001	0.7061	0.0030	0.0000	0.0004	-0.1564	0.6982	0.0073	0.0001	0.0010	0.3164	0.7385
	8	-0.0122	-0.0018	-0.00078	3.2250	-0.0074	-0.0004	-0.0011	0.0000											
	9																			
24	2	0.900	-1.03	0.00	0.05	3.18	0.02	3.17	0.3469	0.7364	0.0032	0.0000	0.0004	-0.0294	0.6871	0.0091	0.0006	0.0013	0.0607	0.6478
	8	-0.0010	0.0002	-0.0002	1.1712	0.0108	0.0001	0.0014	0.0000											
	9																			
24	3	0.898	0.00	0.00	0.05	3.20	0.01	3.18	0.1676	0.6918	0.0033	0.0000	0.0004	-0.0076	0.6859	0.0211	0.0001	0.0028	0.0365	0.6701
	8	-0.0012	0.0002	-0.0002	1.0664	0.0215	0.0001	0.0028	0.0000											
	9																			

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd																			
24	4	7	0.897	0.55	-0.0004	0.05	3.20	0.02	3.19	0.1381	0.6840									
		8	0.0032	-0.0000	0.0004	0.6733	0.0276	0.0007	0.0037	0.0375	0.6748									
		9	-0.0012	0.0002	-0.0002	0.9826	0.0278	0.0002	0.0037											
24	5	7	0.899	1.06	-0.0004	0.05	3.21	0.01	3.19	0.1157	0.6799									
		8	0.0031	0.0000	0.0004	0.6797	0.0335	0.0007	0.0045	0.0263	0.6787									
		9	-0.0010	0.0002	-0.0002	1.1137	0.0329	0.0001	0.0044											
24	6	7	0.898	3.12	-0.0003	0.05	3.24	0.01	3.21	0.1299	0.6387									
		8	0.0023	0.0000	0.0003	0.7472	0.0501	0.0013	0.0064	0.0633	0.6459									
		9	-0.0010	0.0002	-0.0002	1.1604	0.0506	0.0006	0.0065											
24	7	7	0.900	6.22	-0.0002	0.04	3.26	0.02	3.22	0.0877	0.6022									
		8	0.0014	-0.0000	0.0002	0.8252	0.0743	0.0013	0.0089	0.0539	0.5915									
		9	-0.0009	0.0002	-0.0001	0.9818	0.0734	0.0007	0.0086											
24	8	7	0.897	9.31	-0.0002	0.04	3.26	0.01	3.23	0.0535	0.5742									
		8	0.0016	-0.0001	0.0002	0.6685	0.0854	0.0009	0.0098	0.0268	0.5733									
		9	-0.0008	0.0003	-0.0001	0.8396	0.0856	0.0004	0.0098											
24	9	7	0.899	12.45	0.0003	0.04	3.25	0.01	3.23	0.0169	0.5743									
		8	0.0017	-0.0000	0.0003	0.9278	0.0946	0.0003	0.0108	-0.0034	0.5751									
		9	-0.0012	0.0003	-0.0001	0.6735	0.0940	-0.0000	0.0108											
24	10	7	0.897	0.00	-0.0004	0.04	3.20	0.01	3.18	0.1567	0.6821									
		8	0.0034	-0.0000	0.0004	0.6415	0.0218	0.0006	0.0029	0.0342	0.6606									
		9	-0.0009	0.0002	-0.0002	1.2802	0.0229	0.0001	0.0030											
25	1	7	0.898	-3.05	-0.0003	0.04	6.18	0.01	6.24	0.4282	0.7580									
		8	0.0029	-0.0001	0.0003	0.6265	0.0064	0.0005	0.0009	0.0459	0.6709									
		9	-0.0014	0.0002	-0.0003	1.1144	0.0095	0.0000	0.0012											
25	2	7	0.899	-1.00	-0.0004	0.05	6.20	0.02	6.26	0.1182	0.6930									
		8	0.0029	-0.0000	0.0004	0.7902	0.0306	0.0007	0.0042	0.0145	0.6783									
		9	-0.0010	0.0002	-0.0002	1.2025	0.0325	0.0000	0.0044											
25	3	7	0.899	0.03	-0.0004	0.05	6.21	0.01	6.27	0.1108	0.6681									
		8	0.0030	0.0000	0.0004	0.7020	0.0408	0.0009	0.0054	0.0269	0.6755									
		9	-0.0010	0.0002	-0.0002	1.2186	0.0434	0.0002	0.0058											
25	4	7	0.897	0.57	-0.0003	0.05	6.22	0.01	6.27	0.1234	0.6498									
		8	0.0026	0.0000	0.0003	0.7494	0.0446	0.0011	0.0058	0.0487	0.6588									
		9	-0.0011	0.0002	-0.0003	1.4647	0.0480	0.0004	0.0063											

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key →													
25	5	0.898	1.04	-0.00	0.04	6.23	0.01	6.27	0.1238	0.6437					
	6	0.0029	0.0000	0.0003	0.6605	0.482	0.0011	0.0062	0.0486	0.6489					
	9	-0.0011	0.0002	-0.0002	0.9995	0.0504	0.0004	0.0004							
25	6	0.898	3.14	-0.00	0.05	6.24	0.02	6.29	0.1010	0.6166					
	6	0.0023	0.0000	0.0003	0.7919	0.657	0.0013	0.0081	0.0701	0.6085					
	9	-0.0009	0.0002	-0.0002	1.1591	0.0643	0.0009	0.0078							
25	7	0.897	6.23	-0.00	0.04	6.26	0.02	6.30	0.0795	0.5809					
	6	0.0017	-0.0000	0.0002	0.8385	0.617	0.0013	0.0094	0.0556	0.5742					
	9	-0.0005	0.0003	-0.0001	1.6485	0.0808	0.0008	0.0092							
25	8	0.897	9.30	-0.00	0.04	6.25	0.02	6.30	0.0293	0.5756					
	6	0.0015	-0.0000	0.0002	0.9147	0.697	0.0005	0.0103	0.0091	0.5726					
	9	-0.0004	0.0003	-0.0001	1.2858	0.0898	0.0001	0.0102							
25	0	0.899	12.47	0.00	0.04	6.25	0.01	6.29	0.0080	0.5766					
	6	0.0015	-0.0000	0.0002	0.8756	0.696	0.0001	0.0114	-0.0133	0.5753					
	9	-0.0010	0.0002	-0.0002	1.0802	0.0989	-0.0002	0.0113							
25	10	0.900	0.03	-0.00	0.05	6.22	0.01	6.27	0.1054	0.6653					
	6	0.0026	0.0000	0.0004	0.7857	0.412	0.0008	0.0054	0.0239	0.6698					
	9	-0.0015	0.0003	-0.0002	0.9142	0.0439	0.0002	0.0058							
26	3	0.897	-3.02	-0.00	0.04	9.34	0.01	9.44	0.1240	0.7099					
	6	0.0027	-0.0001	0.0004	0.7508	0.291	0.0007	0.0041	0.0124	0.6795					
	9	-0.0013	0.0001	-0.0002	1.0618	0.333	0.0000	0.0045							
26	4	0.898	-0.97	-0.00	0.04	9.36	0.02	9.46	0.1235	0.6524					
	6	0.0027	-0.0000	0.0004	0.8164	0.470	0.0011	0.0061	0.0472	0.6457					
	9	-0.0010	0.0002	-0.0002	1.2392	0.0503	0.0004	0.0065							
26	5	0.899	0.03	-0.00	0.04	9.37	0.01	9.48	0.1192	0.6368					
	6	0.0032	0.0000	0.0004	0.7145	0.563	0.0012	0.0071	0.0583	0.6250					
	9	-0.0011	0.0002	-0.0002	0.9101	0.0569	0.0006	0.0071							
26	6	0.898	0.57	-0.00	0.05	9.37	0.01	9.48	0.1046	0.6284					
	6	0.0029	0.0000	0.0004	0.7889	0.602	0.0012	0.0075	0.0609	0.6169					
	9	-0.0012	0.0002	-0.0002	0.8435	0.0601	0.0007	0.0074							
26	7	0.898	1.06	-0.00	0.05	9.37	0.01	9.49	0.0983	0.6251					
	6	0.0029	0.0000	0.0004	0.6918	0.639	0.0012	0.0079	0.0626	0.6084					
	9	-0.0012	0.0002	-0.0001	0.7707	0.0643	0.0008	0.0078							

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	← See Key →													
26	8	7	0.900	0.0000	5.15	-0.0003	0.0777	0.8328	0.05	9.38	0.0783	0.0012	9.49	0.0813	0.6002
		8	0.0021	0.0000	0.0000	0.0003	0.0683	0.7922	0.0006	0.0769	0.0769	0.0006	0.0094	0.0421	0.5898
		9	-0.0013	0.0002	0.0002	-0.0002	-1.0683						0.0090		
26	9	7	0.896	6.23	-0.0000	0.0002	-0.1183	0.8745	0.02	9.39	0.862	0.0008	9.49	0.0480	0.5779
		8	0.0016	-0.0000	0.0000	0.0002	-1.4628	0.7295	0.0004	0.862	0.862	0.0004	0.0099	0.0230	0.5704
		9	-0.0011	0.0003	-0.0001	-0.0001	-1.4628						0.0099		
26	10	7	0.899	9.31	-0.0000	0.0003	-0.2204	0.9250	0.02	9.39	0.950	0.0003	9.49	0.0194	0.5777
		8	0.0016	-0.0000	0.0000	0.0003	-1.9026	0.7503	-0.0003	0.950	0.950	-0.0003	0.0109	-0.0051	0.5705
		9	-0.0008	0.0003	-0.0001	-0.0001	-1.9026						0.0108	-0.0051	0.5705
26	11	7	0.900	12.47	0.0001	0.0003	-0.3019	0.8035	0.01	9.38	1.031	0.0000	9.49	0.0022	0.5781
		8	0.0018	-0.0001	0.0000	0.0003	-1.4939	0.6374	-0.0003	1.031	1.031	-0.0003	0.0115	-0.0177	0.5724
		9	-0.0009	0.0002	-0.0001	-0.0001	-1.4939						0.0115	-0.0177	0.5724
26	12	7	0.897	0.01	-0.0000	0.0004	0.0051	0.7784	0.01	9.37	0.565	0.0012	9.48	0.1092	0.6324
		8	0.0026	0.0000	0.0000	0.0004	-1.0481	0.7312	0.0006	0.565	0.565	0.0012	0.0071	0.0593	0.6214
		9	-0.0014	0.0003	-0.0002	-0.0002	-1.0481						0.0071	0.0593	0.6214
27	1	7	0.895	-3.01	-0.0001	-0.0003	-0.1818	0.04	0.02	12.25	0.447	0.0010	12.47	0.1173	0.6583
		8	0.0029	-0.0001	0.0000	0.0003	-0.6895	0.6468	0.0004	0.447	0.447	0.0010	0.0058	0.0460	0.6478
		9	-0.0019	0.0002	-0.0002	-0.0002	-0.6895	0.6587	0.0004	0.490	0.490	0.0004	0.0063	0.0460	0.6478
27	2	7	0.899	0.98	-0.0000	0.0004	0.0138	0.04	0.02	12.27	0.621	0.0012	12.48	0.0985	0.6279
		8	0.0026	0.0000	0.0000	0.0004	-1.1408	0.8317	0.0007	0.621	0.621	0.0012	0.0078	0.0563	0.6158
		9	-0.0013	0.0003	-0.0002	-0.0002	-1.1408	0.7906	0.0007	0.636	0.636	0.0007	0.0078	0.0563	0.6158
27	3	7	0.897	0.04	-0.0000	-0.0004	0.0225	0.04	0.02	12.28	0.700	0.0012	12.49	0.0898	0.6174
		8	0.0030	0.0000	0.0000	0.0004	-1.5609	0.9085	0.0007	0.700	0.700	0.0012	0.0086	0.0547	0.6035
		9	-0.0010	0.0003	-0.0001	-0.0001	-1.5609						0.0085	0.0547	0.6035
27	4	7	0.896	0.59	-0.0000	0.0004	0.0352	0.04	0.02	12.28	0.733	0.0012	12.49	0.0867	0.6083
		8	0.0030	0.0000	0.0000	0.0004	-1.1256	0.9198	0.0006	0.733	0.733	0.0012	0.0089	0.0452	0.5978
		9	-0.0012	0.0002	-0.0002	-0.0002	-1.1256						0.0087	0.0452	0.5978
27	5	7	0.895	1.08	-0.0000	-0.0004	0.1348	0.04	0.02	12.28	0.764	0.0012	12.49	0.0825	0.6015
		8	0.0029	0.0000	0.0000	0.0004	-1.1037	0.8018	0.0006	0.764	0.764	0.0012	0.0091	0.0445	0.5942
		9	-0.0013	0.0003	-0.0002	-0.0002	-1.1037						0.0090	0.0445	0.5942
27	6	7	0.898	3.15	-0.0000	-0.0003	-0.0371	0.04	0.02	12.29	0.829	0.0011	12.49	0.0687	0.5803
		8	0.0018	-0.0000	0.0000	0.0003	-1.3910	0.8873	0.0006	0.829	0.829	0.0011	0.0095	0.0400	0.5737
		9	-0.0012	0.0003	-0.0001	-0.0001	-1.3910						0.0095	0.0400	0.5737

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key													
27	7	0.899	6.24	-0.0003	-0.0001	-0.0003	0.04	12.29	0.02	12.49	0.0272	0.5770			
	8	0.0015	-0.0000	0.0003	0.0003	0.0003	0.9417	0.0912	0.0004	0.0105	0.0060	0.5699			
	9	-0.0011	0.0003	-0.0001	-0.0001	-1.7385	0.5490	0.0922	0.0001	0.0105	0.0060				
27	8	0.897	9.33	-0.0001	-0.0002	-2.3065	0.04	12.29	0.02	12.49	0.0093	0.5787			
	9	0.0016	-0.0003	0.0002	0.0002	0.3065	0.7855	0.1010	0.0001	0.0116	0.0136	0.5711			
	9	-0.0009	0.0003	-0.0000	-0.0000	-2.0306	0.4936	0.1003	-0.0002	0.0114	-0.0136				
27	10	0.898	12.47	0.0001	0.0002	-0.5310	0.04	12.28	0.01	12.49	-0.0001	0.5753			
	9	0.0011	-0.0002	0.0002	0.0002	-0.6518	0.2500	0.1035	0.0000	0.0119	-0.0210	0.5698			
	9	-0.0021	0.0002	-0.0002	-0.0002	0.6518	0.6486	0.1028	-0.0004	0.0117	-0.0210				
27	11	0.897	0.04	-0.0004	-0.0002	0.0356	0.04	12.28	0.02	12.49	0.0844	0.6101			
	9	0.0028	0.0000	0.0004	0.0004	0.9992	0.7609	0.0699	0.0011	0.0085	0.0519	0.6040			
	9	-0.0015	0.0003	-0.0002	-0.0002	-0.9992	0.7883	0.0708	0.0007	0.0085	0.0519				
28	1	0.925	-5.14	0.0005	-0.0011	0.4711	-0.59	14.36	-0.61	14.86	0.1507	0.8894			
	9	0.0060	0.0000	0.0000	0.0009	0.0058	0.9183	0.0441	0.0013	0.0078	0.0945	0.6178			
	9	-0.0081	-0.0000	-0.0009	-0.0009	0.0058	0.5595	0.0696	0.0013	0.0086	0.0945				
28	2	0.925	-1.12	0.0006	-0.0011	0.5050	-0.58	14.38	-0.61	14.87	0.1176	0.7945			
	9	0.0061	0.0000	0.0006	0.0011	0.5050	0.9312	0.0590	0.0013	0.0093	0.0804	0.5989			
	9	-0.0080	0.0000	-0.0007	-0.0007	-0.0498	0.4830	0.0827	0.0013	0.0099	0.0804	0.5989			
28	3	0.894	0.04	-0.0003	-0.0002	-0.0048	0.04	15.00	0.02	15.28	0.0753	0.5966			
	9	0.0027	-0.0000	0.0004	0.0004	0.0048	0.7365	0.0794	0.0011	0.0094	0.0403	0.5807			
	9	-0.0018	0.0003	-0.0002	-0.0002	-0.9173	0.8097	0.0799	0.0006	0.0092	0.0403	0.5807			
28	4	0.898	0.60	-0.0003	-0.0002	-1.3661	0.04	15.01	0.02	15.28	0.0740	0.5908			
	9	0.0032	-0.0000	0.0004	0.0004	0.0002	0.7040	0.0814	0.0012	0.0096	0.0417	0.5769			
	9	-0.0013	0.0003	-0.0002	-0.0002	-1.3661	0.8008	0.0816	0.0006	0.0094	0.0417	0.5769			
28	5	0.899	1.08	-0.0003	-0.0002	-1.3787	0.04	15.00	0.02	15.28	0.0736	0.5855			
	9	0.0035	0.0000	0.0004	0.0004	0.0002	0.6584	0.0824	0.0012	0.0096	0.0425	0.5736			
	9	-0.0012	0.0003	-0.0002	-0.0002	-1.3787	0.9041	0.0831	0.0007	0.0095	0.0425	0.5736			
28	6	0.898	3.17	-0.0004	-0.0002	-0.0275	0.04	15.00	0.02	15.27	0.0399	0.5794			
	9	0.0029	-0.0000	0.0004	0.0004	0.0275	0.7082	0.0882	0.0007	0.0102	0.0148	0.5681			
	9	-0.0016	0.0003	-0.0002	-0.0002	-1.0253	0.8489	0.0892	0.0002	0.0101	0.0148	0.5681			
28	7	0.898	6.23	-0.0004	-0.0002	-0.2441	0.04	15.00	0.02	15.28	0.0177	0.5749			
	9	0.0018	-0.0000	0.0002	0.0002	-0.2441	0.7332	0.0964	0.0003	0.0110	0.0177	0.5691			
	9	-0.0006	0.0004	-0.0000	-0.0000	-2.9773	0.6952	0.0971	-0.0000	0.0110	-0.0177	0.5691			

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																																						
28	8	0.899	0.0020	0.0010	0.897	0.0017	0.0010	0.897	0.0030	0.0012	1.249	0.0020	0.0005	1.249	0.0028	0.0005	1.246	0.0026	0.0007	1.248	0.0029	0.0002	1.249	0.0023	0.0002	1.246	0.0012	0.0001	1.248	0.0016	0.0001	1.248	0.0012	0.0001	1.248	0.0016	0.0001			
28	0	0.897	0.0017	0.0010	0.897	0.0017	0.0010	0.897	0.0030	0.0012	1.249	0.0020	0.0005	1.249	0.0028	0.0005	1.246	0.0026	0.0007	1.248	0.0029	0.0002	1.249	0.0023	0.0002	1.246	0.0012	0.0001	1.248	0.0016	0.0001	1.248	0.0016	0.0001	1.248	0.0016	0.0001			
28	10	0.897	0.0030	0.0012	0.897	0.0030	0.0012	0.897	0.0030	0.0012	1.249	0.0020	0.0005	1.249	0.0028	0.0005	1.246	0.0026	0.0007	1.248	0.0029	0.0002	1.249	0.0023	0.0002	1.246	0.0012	0.0001	1.248	0.0016	0.0001	1.248	0.0016	0.0001	1.248	0.0016	0.0001			
29	3	1.249	0.0020	0.0005	1.249	0.0020	0.0005	1.249	0.0020	0.0005	1.249	0.0020	0.0005	1.249	0.0020	0.0005	1.249	0.0020	0.0005	1.249	0.0020	0.0005	1.249	0.0020	0.0005	1.249	0.0020	0.0005	1.249	0.0020	0.0005	1.249	0.0020	0.0005	1.249	0.0020	0.0005	1.249	0.0020	0.0005
29	4	1.249	0.0028	0.0005	1.249	0.0028	0.0005	1.249	0.0028	0.0005	1.249	0.0028	0.0005	1.249	0.0028	0.0005	1.249	0.0028	0.0005	1.249	0.0028	0.0005	1.249	0.0028	0.0005	1.249	0.0028	0.0005	1.249	0.0028	0.0005	1.249	0.0028	0.0005	1.249	0.0028	0.0005	1.249	0.0028	0.0005
29	5	1.246	0.0026	0.0007	1.246	0.0026	0.0007	1.246	0.0026	0.0007	1.246	0.0026	0.0007	1.246	0.0026	0.0007	1.246	0.0026	0.0007	1.246	0.0026	0.0007	1.246	0.0026	0.0007	1.246	0.0026	0.0007	1.246	0.0026	0.0007	1.246	0.0026	0.0007	1.246	0.0026	0.0007	1.246	0.0026	0.0007
29	6	1.248	0.0026	0.0007	1.248	0.0026	0.0007	1.248	0.0026	0.0007	1.248	0.0026	0.0007	1.248	0.0026	0.0007	1.248	0.0026	0.0007	1.248	0.0026	0.0007	1.248	0.0026	0.0007	1.248	0.0026	0.0007	1.248	0.0026	0.0007	1.248	0.0026	0.0007	1.248	0.0026	0.0007	1.248	0.0026	0.0007
29	7	1.248	0.0029	0.0002	1.248	0.0029	0.0002	1.248	0.0029	0.0002	1.248	0.0029	0.0002	1.248	0.0029	0.0002	1.248	0.0029	0.0002	1.248	0.0029	0.0002	1.248	0.0029	0.0002	1.248	0.0029	0.0002	1.248	0.0029	0.0002	1.248	0.0029	0.0002	1.248	0.0029	0.0002	1.248	0.0029	0.0002
29	8	1.249	0.0023	0.0002	1.249	0.0023	0.0002	1.249	0.0023	0.0002	1.249	0.0023	0.0002	1.249	0.0023	0.0002	1.249	0.0023	0.0002	1.249	0.0023	0.0002	1.249	0.0023	0.0002	1.249	0.0023	0.0002	1.249	0.0023	0.0002	1.249	0.0023	0.0002	1.249	0.0023	0.0002	1.249	0.0023	0.0002
29	9	1.246	0.0012	0.0001	1.246	0.0012	0.0001	1.246	0.0012	0.0001	1.246	0.0012	0.0001	1.246	0.0012	0.0001	1.246	0.0012	0.0001	1.246	0.0012	0.0001	1.246	0.0012	0.0001	1.246	0.0012	0.0001	1.246	0.0012	0.0001	1.246	0.0012	0.0001	1.246	0.0012	0.0001	1.246	0.0012	0.0001
29	10	1.248	0.0016	0.0006	1.248	0.0016	0.0006	1.248	0.0016	0.0006	1.248	0.0016	0.0006	1.248	0.0016	0.0006	1.248	0.0016	0.0006	1.248	0.0016	0.0006	1.248	0.0016	0.0006	1.248	0.0016	0.0006	1.248	0.0016	0.0006	1.248	0.0016	0.0006	1.248	0.0016	0.0006	1.248	0.0016	0.0006

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key														
29	11	7	1.249	12.60	-0.0001	-0.000	-0.6430	0.8068	-2.248	0.0849	0.0006	-2.191	-0.0367	0.6433		
		6	0.0014	-0.0001	0.0001	0.0002	-0.8289	0.5804	0.0860	-0.0010	0.0109	0.0110	-0.0610	0.6426		
		5	-0.0011	0.0004	-0.0001	-0.0001	-1.8289	0.5804	0.0860	-0.0010	0.0109	0.0110	-0.0610	0.6426		
29	12	7	1.247	-0.005	-0.0003	-0.000	-0.3239	0.6484	-3.09	0.000	0.02	-2.96	-0.0194	0.7546		
		6	0.0025	-0.0001	0.0003	0.0003	-2.0823	0.6446	-0.0142	0.0002	-0.0021	-0.0021	0.0857	0.7608		
		9	-0.0009	0.0003	-0.0001	-0.0001	-2.0823	0.6446	-0.0142	0.0002	-0.0021	-0.0021	0.0857	0.7608		
30	17	7	1.250	-3.13	0.0002	-0.000	-0.4513	0.7337	-1.07	0.000	0.02	-1.03	-0.0385	0.6790		
		6	0.0018	-0.0001	0.0002	0.0002	1.0424	0.5492	-0.307	0.0000	-0.0042	-0.0042	0.0034	0.6855		
		9	0.0013	0.0002	0.0001	0.0001	1.0424	0.5492	-0.307	0.0000	-0.0042	-0.0042	0.0034	0.6855		
30	18	7	1.250	-1.08	0.0003	-0.000	-0.3921	0.7752	-1.05	0.000	0.02	-1.03	-0.0466	0.7077		
		6	0.0020	-0.0001	0.0003	0.0003	0.8169	0.5429	-0.0128	0.0001	-0.0018	-0.0018	0.0917	0.7040		
		9	0.0018	0.0003	0.0002	0.0002	0.8169	0.5429	-0.0128	0.0001	-0.0018	-0.0018	0.0917	0.7040		
30	19	7	1.249	-0.02	0.0003	-0.000	-0.3535	0.954	-1.04	0.000	0.02	-1.02	-0.1466	0.7215		
		6	0.0023	-0.0001	0.0003	0.0003	0.9842	0.5518	-0.0043	0.0001	-0.0006	-0.0006	0.2570	0.7719		
		9	0.0017	0.0003	0.0001	0.0001	0.9842	0.5518	-0.0043	0.0001	-0.0006	-0.0006	0.2570	0.7719		
30	20	7	1.250	0.52	0.0003	-0.000	-0.3654	0.7146	-1.04	0.000	0.02	-1.01	-1.8985	0.9049		
		6	0.0022	-0.0001	0.0003	0.0003	1.0993	0.6144	-0.0000	0.0002	-0.0001	-0.0001	18.4407	10.5344		
		9	0.0016	0.0003	0.0001	0.0001	1.0993	0.6144	-0.0000	0.0002	-0.0001	-0.0001	18.4407	10.5344		
30	21	7	1.249	1.03	0.0003	-0.000	-0.3393	0.7263	-1.03	0.000	0.02	-1.02	-0.3062	0.5335		
		6	0.0021	-0.0001	0.0003	0.0003	0.9910	0.5734	0.0036	0.0001	0.0003	0.0003	0.3062	0.5335		
		9	0.0018	0.0003	0.0002	0.0002	0.9910	0.5734	0.0036	0.0001	0.0003	0.0003	0.3062	0.5335		
30	22	7	1.250	3.12	0.0003	-0.000	-0.3060	0.6692	-1.01	0.000	0.02	-1.00	-0.0477	0.6692		
		6	0.0019	-0.0001	0.0003	0.0002	1.0887	0.7461	0.0195	0.0002	0.0025	0.0025	0.0517	0.6646		
		9	0.0016	0.0003	0.0002	0.0002	1.0887	0.7461	0.0195	0.0002	0.0025	0.0025	0.0517	0.6646		
30	23	7	1.250	6.28	0.0003	-0.000	-1.0618	0.8663	-0.99	0.000	0.02	-0.99	-0.0127	0.6586		
		6	0.0008	-0.0001	0.0003	0.0003	0.9110	0.7077	0.0468	0.0001	0.0062	0.0062	0.0516	0.6586		
		9	0.0021	0.0003	0.0003	0.0003	0.9110	0.7077	0.0468	0.0001	0.0062	0.0062	0.0516	0.6586		
30	24	7	1.250	9.39	0.0003	-0.000	-1.5072	0.9663	-0.96	0.000	0.02	-0.96	-0.0312	0.6540		
		6	0.0005	-0.0001	0.0003	0.0003	0.8701	1.0963	0.0726	0.0004	0.0095	0.0095	0.0548	0.6540		
		9	0.0021	0.0003	0.0003	0.0003	0.8701	1.0963	0.0726	0.0004	0.0095	0.0095	0.0548	0.6540		
30	25	7	1.249	12.64	0.0004	-0.000	-0.6609	0.8137	-0.94	0.000	0.02	-0.94	-0.0342	0.6379		
		6	0.0012	-0.0001	0.0004	0.0002	1.6879	0.8021	0.0940	0.0006	0.0120	0.0120	0.0628	0.6379		
		9	0.0012	0.0004	0.0002	0.0002	1.6879	0.8021	0.0940	0.0006	0.0120	0.0120	0.0628	0.6379		

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																	
30	25	1.251	-0.001	-0.004	-0.003	-0.3705	0.04	-1.04	0.02	-1.03	-0.1017	0.7812	0.0023	-0.0001	0.0047	0.0001	-0.0008	0.2627	0.8592
	6	0.0015	0.0004	0.0004	0.0002	1.2813	0.6936	-0.0047	-0.0002	-0.0008	0.0008	0.0008							
31	1	1.251	-0.001	-0.003	-0.002	-0.3898	0.04	-0.02	0.01	-0.02	-0.0296	0.6911	0.0021	-0.0001	0.0054	-0.0035	0.0228	0.6946	
	6	0.0012	0.0003	0.0003	0.0001	1.3629	0.6131	-0.0255	-0.0001	-0.0035	0.0001	0.0001							
31	2	1.249	-0.001	-0.005	-0.003	-0.3514	0.04	-0.01	0.01	-0.01	-0.0437	0.7396	0.0014	-0.0003	0.0075	-0.0011	0.1828	0.7516	
	6	0.0014	0.0003	0.0003	0.0001	1.2903	0.5430	-0.0075	-0.0002	-0.0011	0.0001	0.0001							
31	3	1.248	-0.001	-0.003	-0.003	-0.3220	0.04	-0.00	0.02	-0.00	-22.5203	5.8604	0.0026	-0.0001	0.0000	-0.0001	32.1884	15.4762	
	6	0.0017	0.0003	0.0003	0.0001	1.1527	0.5660	-0.0000	-0.0002	-0.0001	0.0001	0.0001							
31	4	1.248	-0.001	-0.005	-0.003	-0.3462	0.04	0.00	0.02	0.00	0.1919	0.6413	0.0022	-0.0001	0.0051	0.0004	0.2409	0.5108	
	6	0.0015	0.0003	0.0003	0.0002	1.2391	0.6393	0.0043	-0.0002	0.0002	-0.0002	0.0002							
31	5	1.248	-0.001	-0.003	-0.003	-0.3103	0.04	0.01	0.01	0.01	0.0990	0.6625	0.0023	-0.0001	0.0098	0.0010	0.0982	0.6216	
	6	0.0016	0.0003	0.0003	0.0001	1.1370	0.5336	0.0081	-0.0001	0.0001	-0.0010	0.0010							
31	9	1.248	-0.001	-0.003	-0.003	-0.3556	0.04	0.01	0.01	0.01	0.1263	0.6752	0.0022	-0.0001	0.0096	0.0011	0.0922	0.6361	
	6	0.0019	0.0003	0.0003	0.0002	0.8302	0.7051	0.0094	-0.0001	0.0001	-0.0011	0.0011							
31	10	1.248	-0.001	-0.003	-0.003	-0.4695	0.04	0.03	0.01	0.02	0.0324	0.6704	0.0020	-0.0001	0.0264	0.0033	0.0503	0.6681	
	6	0.0020	0.0003	0.0003	0.0003	0.7867	0.7484	0.0252	-0.0002	0.0002	-0.0033	0.0033							
31	11	1.250	-0.001	-0.003	-0.004	-1.6489	0.03	0.05	0.02	0.03	-0.0176	0.6605	0.0026	-0.0001	0.0515	0.0070	0.0516	0.6701	
	6	0.0026	0.0003	0.0003	0.0004	0.7014	0.7650	0.0533	-0.0005	0.0005	-0.0070	0.0070							
31	12	1.249	-0.001	-0.003	-0.003	-2.6585	0.03	0.07	0.02	0.07	-0.0321	0.6526	0.0003	-0.0001	0.0773	0.0100	0.0321	0.6505	
	6	0.0018	0.0003	0.0003	0.0003	0.9508	0.8770	0.0778	-0.0008	0.0008	-0.0101	0.0101							
31	13	1.249	-0.001	-0.003	-0.003	-0.5860	0.04	0.09	0.02	0.09	-0.0342	0.6327	0.0019	-0.0001	0.0978	0.0123	0.0342	0.6329	
	6	0.0019	0.0003	0.0003	0.0003	1.0164	0.8041	0.0992	-0.0012	0.0012	-0.0123	0.0123							

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key													
31	14	1.250	-0.0001	-0.0003	-0.3516	0.6913	0.04	0.0001	0.0001	0.0001	-0.0001	-0.0001	-0.0001	-4.9800	3.3818
		0.0023	-0.0004	0.0002	1.3187	0.8640	-0.0005	0.0002	0.0002	0.0002	-0.0002	-0.0002	-2.3212	-0.7716	
32	1	1.248	-0.0001	-0.0001	-0.5175	0.5928	1.02	0.0001	0.0001	0.0001	-0.0001	-0.0001	-0.0215	0.6992	
		0.0016	0.0003	0.0002	1.3269	0.7969	-0.0198	0.0000	0.0000	0.0000	-0.0027	-0.0027	0.0484	0.7049	
32	2	1.249	-0.0001	-0.0003	-0.3503	0.6447	1.05	0.0002	0.0002	0.0002	-0.0004	-0.0004	-0.1730	0.8084	
		0.0024	0.0003	0.0002	1.1978	0.8054	-0.0023	0.0000	0.0000	0.0000	-0.0004	-0.0004	0.6387	0.9542	
32	3	1.250	-0.0001	-0.0002	-0.3760	0.6422	1.06	0.0001	0.0001	0.0001	-0.0005	-0.0005	0.1710	0.6482	
		0.0021	0.0003	0.0002	1.0210	0.6859	0.0054	0.0002	0.0002	0.0002	0.0006	0.0006	-0.2021	0.5482	
32	4	1.250	-0.0001	-0.0002	-0.3660	0.6676	1.06	0.0001	0.0001	0.0001	-0.0013	-0.0013	0.0843	0.6446	
		0.0022	0.0003	0.0002	1.0436	0.5964	0.0094	0.0001	0.0001	0.0001	0.0012	0.0012	-0.0972	0.6528	
32	5	1.250	-0.0001	-0.0002	-0.3381	0.6900	1.07	0.0001	0.0001	0.0001	-0.0018	-0.0018	0.0560	0.6578	
		0.0019	0.0003	0.0002	0.9905	0.6561	0.0138	0.0001	0.0001	0.0001	0.0018	0.0018	-0.0694	0.6534	
32	6	1.250	-0.0001	-0.0002	-0.3733	0.5523	1.09	0.0001	0.0001	0.0001	-0.0018	-0.0018	0.0099	0.6682	
		0.0016	0.0003	0.0002	0.9442	0.6140	0.0307	0.0003	0.0003	0.0003	0.0041	0.0041	-0.0545	0.6665	
32	7	1.248	-0.0001	-0.0002	-0.7612	0.5467	1.11	0.0001	0.0001	0.0001	-0.0018	-0.0018	0.0269	0.6616	
		0.0018	0.0003	0.0002	1.0581	0.6798	0.0584	0.0006	0.0006	0.0006	0.0077	0.0077	-0.0545	0.6601	
32	8	1.248	-0.0001	-0.0002	-2.0672	0.8560	1.14	0.0001	0.0001	0.0001	-0.0018	-0.0018	0.0343	0.6471	
		0.0004	0.0003	0.0003	0.8455	0.7115	0.0826	0.0009	0.0009	0.0009	0.0106	0.0106	-0.0582	0.6471	
32	9	1.249	-0.0001	-0.0002	-0.8254	0.8860	1.16	0.0001	0.0001	0.0001	-0.0018	-0.0018	0.0362	0.6228	
		0.0010	0.0004	0.0002	1.4189	0.7967	0.1032	0.0012	0.0012	0.0012	0.0128	0.0128	-0.0627	0.6228	
32	10	1.249	-0.0001	-0.0002	-0.3616	0.6461	1.05	0.0001	0.0001	0.0001	-0.0018	-0.0018	0.1361	0.6011	
		0.0023	0.0004	0.0002	1.2005	0.6526	0.0052	0.0002	0.0002	0.0002	0.0005	0.0005	-0.2289	0.5416	

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key															
32	11	7	1.250	-3.10	-0.0001	-0.0002	-0.0001	-0.4523	0.5632	0.04	-0.0083	5.15	0.02	0.0000	-0.0012	3.14	-0.0253	0.7202
		8	0.0018	-0.0003	0.0001	0.0002	0.0001	1.2021	0.5900	0.04	-0.0083	-0.0085	0.0000	-0.0002	-0.0012	-0.0012	0.1640	0.7148
		9	0.0014	0.0003	0.0001	0.0002	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001
33	1	7	1.249	-1.05	-0.0001	-0.0002	-0.0001	-0.3507	0.04	0.04	3.17	0.02	0.0001	0.0001	3.16	0.0620	0.6846	
		8	0.0024	-0.0003	0.0001	0.0002	0.0001	1.3391	0.6163	0.04	0.0077	0.0073	0.0001	0.0002	0.0001	0.0001	0.1825	0.6384
		9	0.0014	0.0003	0.0001	0.0002	0.0001	0.0001	0.7810	0.04	0.0073	-0.0002	0.0001	0.0002	0.0001	0.0001	-0.0001	0.6846
33	2	7	1.249	-3.09	-0.0001	-0.0002	-0.0001	-0.4171	0.04	0.04	3.14	0.02	0.0000	-0.0012	3.14	-0.0206	0.7249	
		8	0.0020	-0.0001	0.0001	0.0002	0.0001	1.3612	0.5841	0.04	-0.0083	-0.0089	0.0000	-0.0002	-0.0012	0.1576	0.7108	
		9	0.0012	0.0003	0.0001	0.0002	0.0001	0.0001	0.5918	0.04	-0.0083	-0.0089	0.0000	-0.0002	-0.0012	0.1576	0.7108	
33	3	7	1.250	-0.00	-0.0001	-0.0002	-0.0001	-0.3302	0.04	0.04	3.18	0.02	0.0000	0.0000	3.16	0.0291	0.6608	
		8	0.0026	-0.0001	0.0001	0.0002	0.0001	1.2020	0.5496	0.04	0.0170	0.0152	0.0000	0.0002	0.0022	0.0291	0.6608	
		9	0.0016	0.0003	0.0001	0.0002	0.0001	0.0001	0.6430	0.04	0.0152	-0.0002	0.0000	0.0002	0.0019	-0.0097	0.6553	
33	4	7	1.249	0.55	-0.0001	-0.0002	-0.0001	-0.3267	0.04	0.04	3.19	0.02	0.0001	0.0001	3.17	0.0251	0.6640	
		8	0.0023	-0.0001	0.0001	0.0002	0.0001	1.1355	0.6250	0.04	0.0218	0.0207	0.0001	0.0002	0.0027	0.0251	0.6640	
		9	0.0017	0.0003	0.0001	0.0002	0.0001	0.0001	0.5925	0.04	0.0207	-0.0002	0.0000	0.0002	0.0027	-0.0091	0.6678	
33	5	7	1.249	1.05	-0.0001	-0.0002	-0.0001	-0.3296	0.04	0.04	3.20	0.02	0.0000	0.0000	3.17	0.0158	0.6686	
		8	0.0022	-0.0001	0.0001	0.0002	0.0001	1.1510	0.6145	0.04	0.0268	0.0246	0.0000	0.0002	0.0035	0.0158	0.6686	
		9	0.0017	0.0003	0.0001	0.0002	0.0001	0.0001	0.6242	0.04	0.0246	-0.0002	0.0000	0.0002	0.0033	-0.0096	0.6742	
33	6	7	1.249	3.14	-0.0001	-0.0002	-0.0001	-0.3733	0.04	0.04	3.22	0.02	0.0001	0.0001	3.18	-0.0128	0.6687	
		8	0.0016	-0.0001	0.0001	0.0002	0.0001	1.0509	0.5523	0.04	0.0447	0.0416	0.0001	0.0004	0.0059	-0.0128	0.6746	
		9	0.0018	0.0003	0.0001	0.0002	0.0001	0.0001	0.6384	0.04	0.0416	-0.0002	0.0000	0.0004	0.0056	-0.0058	0.6746	
33	7	7	1.248	6.27	-0.0002	-0.0003	-0.0001	-1.2990	0.03	0.03	3.23	0.02	0.0004	0.0007	3.20	-0.0345	0.6580	
		8	0.0007	-0.0002	0.0001	0.0002	0.0001	0.8026	0.6363	0.03	0.0692	0.0690	0.0000	0.0007	0.0091	-0.0345	0.6599	
		9	0.0025	0.0004	0.0001	0.0002	0.0001	0.0001	0.6096	0.03	0.0690	-0.0002	0.0000	0.0007	0.0091	-0.0345	0.6599	
33	8	7	1.249	9.41	-0.0001	-0.0002	-0.0001	-1.7540	0.04	0.04	3.25	0.02	0.0006	0.0006	3.36	-0.0363	0.6405	
		8	0.0005	-0.0001	0.0001	0.0002	0.0001	0.8840	1.1790	0.04	0.0922	0.0477	0.0006	0.0006	0.0118	-0.0363	1.7056	
		9	0.0022	0.0003	0.0001	0.0002	0.0001	0.0001	0.6750	0.04	0.0477	-0.0002	0.0002	0.0006	0.0163	0.6515	0.6405	
33	9	7	1.249	12.66	-0.0001	-0.0002	-0.0001	-1.2695	0.04	0.04	3.27	0.02	0.0008	0.0013	3.22	-0.0394	0.6220	
		8	0.0006	-0.0001	0.0001	0.0002	0.0001	1.5719	0.9024	0.04	0.1102	-0.0002	0.0008	0.0013	0.0137	-0.0394	0.6220	
		9	0.0013	0.0004	0.0001	0.0002	0.0001	0.0001	0.6672	0.04	0.1102	-0.0002	0.0008	0.0013	0.0136	-0.0394	0.6193	
33	10	7	1.250	-0.03	-0.0001	-0.0002	-0.0001	-0.3812	0.04	0.04	3.18	0.02	0.0000	0.0000	3.17	0.0270	0.6599	
		8	0.0022	-0.0001	0.0001	0.0002	0.0001	1.2174	0.6283	0.04	0.0169	0.0154	0.0000	0.0000	0.0022	0.0270	0.6599	
		9	0.0017	0.0004	0.0001	0.0002	0.0001	0.0001	0.5882	0.04	0.0154	-0.0002	0.0000	0.0000	0.0020	-0.0063	0.6599	

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																	
35	4	1.250	-1.00	-0.00	0.04	9.34	0.02	9.45	-0.0115	0.7042	0.0001	0.0003	-0.4189	0.7448	0.0395	-0.0000	0.0055	-0.0716	0.6921
	8	0.0020	-0.0001	0.0002	0.7448	0.0395	0.0000	0.0055	-0.0115	0.7042	0.0001	0.0003	-0.4189	0.7448	0.0395	-0.0000	0.0055	-0.0716	0.6921
	9	0.0015	0.0003	0.0002	0.7025	0.0408	-0.0005	0.0056	-0.0716	0.7025	0.0408	-0.0005	0.9795	0.7025	0.0408	-0.0005	0.0056	-0.0716	0.6921
35	5	1.249	0.03	-0.00	0.04	9.35	0.02	9.47	-0.0178	0.6919	0.0001	0.0003	-0.3454	0.7311	0.0485	-0.0001	0.0067	-0.0637	0.6832
	8	0.0022	-0.0001	0.0002	0.7311	0.0485	-0.0001	0.0067	-0.0637	0.7311	0.0485	-0.0001	0.3454	0.7311	0.0485	-0.0001	0.0067	-0.0637	0.6832
	9	0.0016	0.0003	0.0002	0.7959	0.0491	-0.0006	0.0067	-0.0637	0.7959	0.0491	-0.0006	1.0121	0.7959	0.0491	-0.0006	0.0067	-0.0637	0.6832
35	6	1.249	0.61	-0.00	0.04	9.36	0.02	9.46	-0.0202	0.6887	0.0001	0.0003	-0.3497	0.7096	0.0537	-0.0002	0.0074	-0.0634	0.6744
	8	0.0021	-0.0001	0.0002	0.7096	0.0537	-0.0002	0.0074	-0.0634	0.7096	0.0537	-0.0002	1.1616	0.7096	0.0537	-0.0002	0.0074	-0.0634	0.6744
	9	0.0015	0.0003	0.0002	0.8981	0.0540	-0.0006	0.0072	-0.0634	0.8981	0.0540	-0.0006	-0.3497	0.8981	0.0540	-0.0006	0.0072	-0.0634	0.6744
35	7	1.249	1.09	-0.00	0.04	9.36	0.02	9.47	-0.0248	0.6829	0.0001	0.0003	-0.3301	0.6793	0.0585	-0.0002	0.0079	-0.0622	0.6735
	8	0.0022	-0.0001	0.0002	0.6793	0.0585	-0.0002	0.0079	-0.0622	0.6793	0.0585	-0.0002	1.1088	0.6793	0.0585	-0.0002	0.0079	-0.0622	0.6735
	9	0.0015	0.0003	0.0002	0.8646	0.0583	-0.0007	0.0078	-0.0622	0.8646	0.0583	-0.0007	-0.3301	0.8646	0.0583	-0.0007	0.0078	-0.0622	0.6735
35	8	1.249	3.20	-0.00	0.04	9.38	0.02	9.48	-0.0290	0.6686	0.0001	0.0003	-0.4599	0.6308	0.0750	-0.0004	0.0100	-0.0614	0.6594
	8	0.0014	-0.0001	0.0003	0.6308	0.0750	-0.0004	0.0100	-0.0614	0.6308	0.0750	-0.0004	0.7274	0.6308	0.0750	-0.0004	0.0100	-0.0614	0.6594
	9	0.0022	0.0003	0.0003	0.6763	0.0745	-0.0009	0.0098	-0.0614	0.6763	0.0745	-0.0009	-0.4599	0.6763	0.0745	-0.0009	0.0098	-0.0614	0.6594
35	9	1.249	6.34	-0.00	0.04	9.40	0.02	9.49	-0.0337	0.6351	0.0001	0.0003	-1.1191	0.604	0.0961	-0.0012	0.0124	-0.0613	0.6447
	8	0.0008	-0.0001	0.0004	0.604	0.0961	-0.0012	0.0124	-0.0613	0.604	0.0961	-0.0012	0.7520	0.604	0.0961	-0.0012	0.0124	-0.0613	0.6447
	9	0.0025	0.0003	0.0004	0.8322	0.0982	-0.0013	0.0124	-0.0613	0.8322	0.0982	-0.0013	-1.1191	0.8322	0.0982	-0.0013	0.0124	-0.0613	0.6447
35	10	1.250	9.47	-0.00	0.03	9.42	0.02	9.50	-0.0373	0.6223	0.0001	0.0003	-2.9415	0.5901	0.1147	-0.0008	0.0142	-0.0586	0.6153
	8	0.0003	-0.0001	0.0003	0.5901	0.1147	-0.0008	0.0142	-0.0586	0.5901	0.1147	-0.0008	1.0819	0.5901	0.1147	-0.0008	0.0142	-0.0586	0.6153
	9	0.0016	0.0003	0.0003	1.0148	0.1156	-0.0013	0.0142	-0.0586	1.0148	0.1156	-0.0013	-2.9415	1.0148	0.1156	-0.0013	0.0142	-0.0586	0.6153
35	11	1.250	12.65	-0.00	0.04	9.42	0.02	9.50	-0.0431	0.6131	0.0001	0.0003	-0.5436	0.5909	0.1285	-0.0011	0.0157	-0.0645	0.6060
	8	0.0015	-0.0001	0.0002	0.5909	0.1285	-0.0011	0.0157	-0.0645	0.5909	0.1285	-0.0011	1.3042	0.5909	0.1285	-0.0011	0.0157	-0.0645	0.6060
	9	0.0015	0.0004	0.0002	0.9372	0.1283	-0.0016	0.0155	-0.0645	0.9372	0.1283	-0.0016	-0.5436	0.9372	0.1283	-0.0016	0.0155	-0.0645	0.6060
35	12	1.249	0.04	-0.00	0.04	9.35	0.03	9.46	-0.0233	0.6819	0.0001	0.0003	-0.3226	0.6593	0.0492	-0.0002	0.0066	-0.0648	0.6792
	8	0.0025	-0.0001	0.0002	0.6593	0.0492	-0.0002	0.0066	-0.0648	0.6593	0.0492	-0.0002	1.1465	0.6593	0.0492	-0.0002	0.0066	-0.0648	0.6792
	9	0.0016	0.0003	0.0002	0.7786	0.0492	-0.0002	0.0066	-0.0648	0.7786	0.0492	-0.0002	-0.3226	0.7786	0.0492	-0.0002	0.0066	-0.0648	0.6792
36	1	1.252	-3.01	-0.00	0.04	12.23	0.02	12.43	-0.0197	0.7027	0.0001	0.0003	-0.3846	0.7299	0.0369	-0.0001	0.0053	-0.0758	0.6890
	8	0.0020	-0.0001	0.0002	0.7299	0.0369	-0.0001	0.0053	-0.0758	0.7299	0.0369	-0.0001	1.3448	0.7299	0.0369	-0.0001	0.0053	-0.0758	0.6890
	9	0.0012	0.0003	0.0001	0.7027	0.0389	-0.0005	0.0053	-0.0758	0.7027	0.0389	-0.0005	-0.3846	0.7027	0.0389	-0.0005	0.0053	-0.0758	0.6890
36	2	1.247	-0.98	-0.00	0.04	12.25	0.01	12.44	-0.0302	0.6866	0.0001	0.0003	-0.3064	0.6199	0.0551	-0.0003	0.0077	-0.0725	0.6807
	8	0.0027	-0.0001	0.0002	0.6199	0.0551	-0.0003	0.0077	-0.0725	0.6199	0.0551	-0.0003	1.2785	0.6199	0.0551	-0.0003	0.0077	-0.0725	0.6807
	9	0.0014	0.0003	0.0002	0.9197	0.0566	-0.0008	0.0077	-0.0725	0.9197	0.0566	-0.0008	-0.3064	0.9197	0.0566	-0.0008	0.0077	-0.0725	0.6807

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd																						
36	5	7	1.250	0.0023	0.0017	0.06	-0.0001	0.0003	-0.0003	0.0002	-0.0003	0.04	0.7261	0.0639	0.0652	12.26	0.003	-0.0003	0.02	0.0003	0.0086	-0.0311	0.6771
		8										0.7281	0.0652	0.0652	12.45	0.0008	-0.0008	0.0008	-0.0008	0.0087	-0.0654	0.6733	
		9										0.7281	0.0652	0.0652	12.45	0.0008	-0.0008	0.0008	-0.0008	0.0087	-0.0654	0.6733	
36	4	7	1.249	0.0022	0.0018	0.62	-0.0001	0.0003	-0.0003	0.0002	-0.0003	0.04	0.7056	0.0691	0.0699	12.27	0.003	-0.0003	0.02	0.0004	0.0092	-0.0323	0.6720
		8										0.7056	0.0691	0.0699	12.45	0.0008	-0.0008	0.0008	-0.0008	0.0094	-0.0638	0.6726	
		9										0.7056	0.0691	0.0699	12.45	0.0008	-0.0008	0.0008	-0.0008	0.0094	-0.0638	0.6726	
36	5	7	1.250	0.0020	0.0022	1.12	-0.0001	0.0003	-0.0003	0.0002	-0.0003	0.04	0.7542	0.0733	0.0742	12.28	0.003	-0.0003	0.03	0.0004	0.0097	-0.0337	0.6679
		8										0.7542	0.0733	0.0742	12.46	0.0009	-0.0009	0.0009	-0.0009	0.0099	-0.0626	0.6688	
		9										0.7542	0.0733	0.0742	12.46	0.0009	-0.0009	0.0009	-0.0009	0.0099	-0.0626	0.6688	
36	6	7	1.248	0.0017	0.0017	3.22	-0.0001	0.0003	-0.0002	0.0002	-0.0002	0.04	0.6726	0.0894	0.0891	12.29	0.002	-0.0002	0.02	0.0006	0.0116	-0.0348	0.6510
		8										0.6726	0.0894	0.0891	12.47	0.0011	-0.0011	0.0011	-0.0011	0.0115	-0.0642	0.6474	
		9										0.6726	0.0894	0.0891	12.47	0.0011	-0.0011	0.0011	-0.0011	0.0115	-0.0642	0.6474	
36	7	7	1.248	0.0007	0.0025	6.35	-0.0001	0.0004	-0.0001	0.0003	-0.0001	0.04	0.9998	0.1076	0.1100	12.30	0.002	-0.0002	0.02	0.0007	0.0135	-0.0343	0.6319
		8										0.9998	0.1076	0.1100	12.47	0.0012	-0.0012	0.0012	-0.0012	0.0137	-0.0587	0.6226	
		9										0.9998	0.1076	0.1100	12.47	0.0012	-0.0012	0.0012	-0.0012	0.0137	-0.0587	0.6226	
36	8	7	1.249	0.0002	0.0020	9.46	-0.0001	0.0003	-0.0001	0.0003	-0.0001	0.04	1.9481	0.1243	0.1250	12.32	0.002	-0.0002	0.02	0.0010	0.0152	-0.0423	0.6156
		8										1.9481	0.1243	0.1250	12.49	0.0015	-0.0015	0.0015	-0.0015	0.0152	-0.0628	0.6100	
		9										1.9481	0.1243	0.1250	12.49	0.0015	-0.0015	0.0015	-0.0015	0.0152	-0.0628	0.6100	
36	9	7	1.249	0.0010	0.0014	12.67	-0.0001	0.0004	-0.0002	0.0002	-0.0002	0.05	1.0263	0.1329	0.1358	12.33	0.002	-0.0002	0.02	0.0010	0.0160	-0.0397	0.6036
		8										1.0263	0.1329	0.1358	12.49	0.0016	-0.0016	0.0016	-0.0016	0.0160	-0.0630	0.6000	
		9										1.0263	0.1329	0.1358	12.49	0.0016	-0.0016	0.0016	-0.0016	0.0160	-0.0630	0.6000	
36	10	7	1.247	0.0023	0.0014	0.06	-0.0001	0.0004	-0.0002	0.0002	-0.0002	0.04	0.6822	0.0643	0.0648	12.26	0.003	-0.0003	0.03	0.0004	0.0087	-0.0346	0.6727
		8										0.6822	0.0643	0.0648	12.45	0.0008	-0.0008	0.0008	-0.0008	0.0087	-0.0669	0.6731	
		9										0.6822	0.0643	0.0648	12.45	0.0008	-0.0008	0.0008	-0.0008	0.0087	-0.0669	0.6731	
37	1	7	1.249	0.0019	0.0013	-2.99	-0.0001	0.0003	-0.0001	0.0001	-0.0001	0.04	0.6165	0.0514	0.0544	14.96	0.002	-0.0002	0.01	0.0003	0.0071	-0.0359	0.6903
		8										0.6165	0.0514	0.0544	15.24	0.0008	-0.0008	0.0008	-0.0008	0.0074	-0.0746	0.6827	
		9										0.6165	0.0514	0.0544	15.24	0.0008	-0.0008	0.0008	-0.0008	0.0074	-0.0746	0.6827	
37	2	7	1.249	0.0022	0.0019	-0.94	-0.0001	0.0003	-0.0002	0.0002	-0.0002	0.04	0.7468	0.0694	0.0717	14.97	0.002	-0.0002	0.02	0.0005	0.0093	-0.0377	0.6730
		8										0.7468	0.0694	0.0717	15.25	0.0010	-0.0010	0.0010	-0.0010	0.0096	-0.0700	0.6709	
		9										0.7468	0.0694	0.0717	15.25	0.0010	-0.0010	0.0010	-0.0010	0.0096	-0.0700	0.6709	
37	3	7	1.248	0.0025	0.0021	0.06	-0.0001	0.0003	-0.0003	0.0002	-0.0003	0.04	0.6458	0.0775	0.0795	14.98	0.002	-0.0002	0.02	0.0005	0.0103	-0.0356	0.6661
		8										0.6458	0.0775	0.0795	15.26	0.0010	-0.0010	0.0010	-0.0010	0.0104	-0.0652	0.6591	
		9										0.6458	0.0775	0.0795	15.26	0.0010	-0.0010	0.0010	-0.0010	0.0104	-0.0652	0.6591	

Run Pt. Cd

See Key

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key															
37	4	7	1.248	0.64	-0.0003	-0.0002	-0.0003	0.05	14.99	0.0821	0.0005	15.26	-0.0348	0.6627				
	8	9	0.0022	-0.0001	0.0002	0.0002	0.6698	0.0839	-0.0010	0.0110	-0.0635	0.6570						
	9	7	0.0021	0.0003	0.0002	0.0002	0.5144	0.0839	0.0010	0.0110	-0.0635	0.6570						
37	5	7	1.249	1.12	-0.0002	-0.0002	0.04	14.99	0.0859	0.0006	15.26	-0.0359	0.6552					
	8	9	0.0021	-0.0001	0.0002	0.0002	0.6783	0.0875	-0.0011	0.0113	-0.0635	0.6507						
	9	7	0.0023	0.0003	0.0002	0.0002	0.5592	0.0875	-0.0011	0.0113	-0.0635	0.6507						
37	6	7	1.249	3.23	-0.0001	-0.0002	0.05	15.01	0.1001	0.0007	15.26	-0.0358	0.6397					
	8	9	0.0015	-0.0001	0.0002	0.0002	0.6339	0.1014	-0.0013	0.0128	-0.0641	0.6353						
	9	7	0.0023	0.0003	0.0002	0.0002	0.5039	0.1014	-0.0013	0.0128	-0.0641	0.6353						
37	7	7	1.248	6.36	-0.0001	-0.0003	0.04	15.02	0.1168	0.0008	15.28	-0.0376	0.6239					
	8	9	0.0008	-0.0001	0.0003	0.0003	0.9057	0.1179	-0.0012	0.0144	-0.0524	0.6114						
	9	7	0.0026	0.0003	0.0003	0.0003	0.6484	0.1179	-0.0012	0.0144	-0.0524	0.6114						
37	8	7	1.248	9.47	-0.0001	-0.0002	0.05	15.03	0.1319	0.0011	15.28	-0.0435	0.6116					
	8	9	0.0005	-0.0001	0.0002	0.0002	1.0649	0.1328	-0.0016	0.0160	-0.0637	0.6048						
	9	7	0.0021	0.0003	0.0002	0.0002	0.6747	0.1328	-0.0016	0.0160	-0.0637	0.6048						
37	9	7	1.249	12.67	-0.0001	-0.0001	0.04	15.04	0.1361	0.0017	15.28	-0.0411	0.5983					
	8	9	0.0010	-0.0001	0.0001	0.0001	0.9176	0.1350	-0.0011	0.0161	-0.0634	0.5936						
	9	7	0.0016	0.0003	0.0001	0.0001	0.6020	0.1361	-0.0017	0.0161	-0.0634	0.5936						
37	10	7	1.250	0.09	-0.0001	-0.0002	0.04	14.98	0.0802	0.0005	15.25	-0.0375	0.6614					
	8	9	0.0022	-0.0001	0.0002	0.0002	0.6606	0.0802	-0.0010	0.0105	-0.0650	0.6599						
	9	7	0.0024	0.0003	0.0002	0.0002	0.4984	0.0802	-0.0010	0.0105	-0.0650	0.6599						
38	10	7	1.048	-3.13	-0.0004	-0.0001	0.05	-3.13	0.0005	0.0001	-2.99	-0.0487	0.6651					
	8	9	0.0033	-0.0002	0.0001	0.0001	0.6529	-0.0514	0.0005	-0.0068	-0.0112	0.6655						
	9	7	0.0022	0.0001	0.0001	0.0001	0.3197	-0.0497	0.0001	-0.0066	-0.0112	0.6655						
38	11	7	1.047	-1.06	-0.0005	-0.0001	0.04	-3.11	0.0002	0.0001	-2.98	-0.0486	0.6750					
	8	9	0.0040	-0.0002	0.0002	0.0002	0.7184	-0.0274	0.0002	-0.0037	-0.0214	0.6757						
	9	7	0.0020	0.0002	0.0001	0.0001	0.3994	-0.0271	-0.0001	-0.0036	-0.0214	0.6757						
38	12	7	1.048	-0.05	-0.0006	-0.0002	0.05	-3.10	0.0002	0.0002	-2.98	-0.0685	0.6848					
	8	9	0.0045	-0.0002	0.0002	0.0002	0.6951	-0.0170	0.0002	-0.0023	-0.0616	0.6956						
	9	7	0.0027	0.0002	0.0002	0.0002	0.5208	-0.0171	-0.0002	-0.0023	-0.0616	0.6956						
38	13	7	1.048	0.51	-0.0005	-0.0002	0.05	-3.09	0.0001	0.0002	-2.96	-0.0637	0.7011					
	8	9	0.0041	-0.0002	0.0002	0.0002	0.7134	-0.0118	0.0001	-0.0015	-0.1056	0.7011						
	9	7	0.0024	0.0002	0.0002	0.0002	0.4699	-0.0110	-0.0002	-0.0015	-0.1056	0.7011						

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key →																					
38	14	1.049	0.0048	0.0025	0.0002	0.0002	0.0002	0.0006	-0.0002	-0.2179	0.5460	0.0047	0.5588	-0.0072	-0.0066	0.0002	0.0002	-0.0010	-0.0010	-0.1694	0.2048	0.7165	0.7602
38	15	1.050	0.0039	0.0024	0.0002	0.0002	0.0005	0.0002	0.0002	-0.2287	0.6145	0.0534	0.5973	0.0145	0.0118	0.0003	0.0001	0.0016	0.0014	0.1384	-0.0768	0.6549	0.5975
38	16	1.050	0.0029	0.0035	0.0003	0.0003	0.0004	0.0003	0.0003	-0.2989	0.4365	0.0599	0.5161	0.0444	0.0471	0.0003	0.0003	0.0058	0.0061	0.0120	-0.0394	0.6627	0.6545
38	17	1.047	0.0022	0.0026	0.0003	0.0003	0.0003	0.0003	0.0003	-0.3756	0.6230	0.7896	0.7153	0.0752	0.0784	0.0006	0.0006	0.0098	0.0102	-0.0049	-0.0391	0.6559	0.6538
38	18	1.049	0.0020	0.0026	0.0003	0.0003	0.0003	0.0003	0.0003	-0.3418	0.5700	0.7722	0.6455	0.1026	0.1056	0.0009	0.0009	0.0130	0.0134	-0.0124	-0.0427	0.6372	0.6356
38	19	1.048	0.0043	0.0022	0.0002	0.0002	0.0005	0.0002	0.0002	-0.2643	0.6562	0.6895	0.5660	-0.0172	-0.0175	0.0002	0.0002	0.0024	0.0025	-0.0580	-0.0631	0.7117	0.7143
39	1	1.047	0.0036	0.0017	0.0002	0.0002	0.0004	0.0001	0.0001	-0.3340	0.7980	0.6665	0.5421	-0.0367	-0.0363	0.0003	0.0003	0.0049	0.0049	-0.0442	-0.0062	0.6737	0.6778
39	2	1.046	0.0038	0.0018	0.0002	0.0002	0.0005	0.0002	0.0002	-0.3055	0.7832	0.6859	0.5883	-0.0146	-0.0135	0.0002	0.0002	0.0019	0.0019	-0.0666	-0.0914	0.6729	0.7068
39	3	1.050	0.0043	0.0021	0.0002	0.0002	0.0005	0.0002	0.0002	-0.2536	0.6790	0.6837	0.5798	-0.0059	-0.0048	0.0002	0.0002	0.0008	0.0007	-0.1959	-0.2483	0.7087	0.7349
39	4	1.048	0.0043	0.0023	0.0003	0.0003	0.0006	0.0002	0.0002	-0.2423	0.6496	0.7059	0.6092	-0.0005	-0.0007	0.0002	0.0002	0.0000	0.0000	-2.5094	-1.5416	0.6972	-0.0527
39	5	1.048	0.0042	0.0022	0.0003	0.0003	0.0005	0.0002	0.0002	-0.2419	0.6985	0.6992	0.6612	-0.0035	-0.0043	0.0001	0.0001	0.0004	0.0004	0.4151	-0.2262	0.6541	0.5591

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																				
39	6	1.048	0.0038	0.0024	5.12	-0.0001	0.0003	-0.0004	0.0003	-0.2395	0.6433	0.6081	-1.00	0.255	0.0243	0.0002	0.0002	-0.0033	0.0031	-0.0516	0.6642	0.6378
39	0	1.048	0.0033	0.0033	6.23	-0.0001	0.0003	-0.0004	0.0004	-0.2761	0.6356	0.6706	-0.97	0.587	0.0606	0.0000	0.0005	-0.0097	0.0080	-0.0017	0.6650	0.6620
39	8	1.049	0.0024	0.0031	9.36	-0.0001	0.0003	-0.0003	0.0004	-0.3308	0.7321	0.6801	-0.94	0.881	0.0907	0.0002	0.0007	-0.0096	0.0118	-0.0116	0.6511	0.6509
39	0	1.049	0.0021	0.0024	12.56	-0.0001	0.0003	-0.0003	0.0003	-0.2834	0.8088	0.7629	-0.92	1.104	0.1130	0.0001	0.0008	-0.0093	0.0138	-0.0083	0.6287	0.6270
39	10	1.049	0.0041	0.0019	-0.05	-0.0002	0.0003	-0.0005	0.0002	-0.2815	0.6635	0.6202	-1.03	-0.053	0.0038	0.0002	0.0002	-0.0001	0.0006	-0.0006	0.7705	0.8786
40	1	1.049	0.0033	0.0017	3.12	-0.0002	0.0002	-0.0004	0.0001	-0.3727	0.6410	0.4877	-0.03	-0.302	0.0000	0.0002	0.0001	-0.0002	0.0040	-0.0437	0.6732	0.6837
40	2	1.052	0.0041	0.0022	-1.10	-0.0002	0.0002	-0.0005	0.0002	-0.2925	0.6697	0.5378	-0.01	-0.100	0.0089	0.0002	0.0002	-0.0013	0.0013	-0.1097	0.6709	0.7353
40	3	1.047	0.0042	0.0024	-0.03	-0.0002	0.0002	-0.0005	0.0002	-0.2479	0.6866	0.5582	-0.01	-0.001	0.0009	0.0003	0.0002	-0.0000	0.0000	-0.6549	0.3499	0.1081
40	4	1.049	0.0041	0.0021	0.52	-0.0002	0.0003	-0.0005	0.0002	-0.2645	0.6657	0.5606	0.00	0.048	0.0000	0.0003	0.0001	-0.0006	0.0006	0.3239	0.7206	0.5958
40	5	1.047	0.0040	0.0019	1.01	-0.0001	0.0003	-0.0005	0.0002	-0.2409	0.6676	0.6592	0.00	0.097	0.0000	0.0003	0.0001	0.0013	0.0012	0.1592	0.6869	0.6118
40	6	1.049	0.0036	0.0025	5.12	-0.0001	0.0003	-0.0004	0.0003	-0.2506	0.6341	0.6812	0.02	0.320	0.0315	0.0001	0.0003	0.0041	0.0041	0.0290	0.6711	0.6545

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key														
40	7	1.050	0.0025	0.0028	6.24	-0.0001	0.0003	-0.3573	0.7351	0.05	0.0638	-0.0000	0.02	0.04	-0.0058	0.6552
	8	0.0024	0.0028	0.0003	0.0003	0.0003	0.0003	0.5919	0.6338	0.0666	-0.0005	0.0005	0.0005	0.0007	-0.0422	0.6599
40	7	1.049	0.0024	0.0028	9.39	-0.0001	0.0003	-0.3221	0.7382	0.07	0.0926	-0.0002	0.02	0.05	-0.0122	0.6472
	8	0.0024	0.0028	0.0003	0.0003	0.0003	0.0003	0.5947	0.6983	0.0958	-0.0008	0.0008	0.0008	0.0123	-0.0433	0.6464
40	7	1.047	0.0018	0.0020	12.56	-0.0001	0.0003	-0.3495	0.7770	0.10	0.1132	-0.0002	0.02	0.07	-0.0038	0.6252
	8	0.0018	0.0020	0.0003	0.0003	0.0002	0.0002	0.8507	0.7263	0.1152	-0.0007	0.0007	0.0007	0.0142	-0.0316	0.6180
40	7	1.047	0.0042	0.0021	-0.08	-0.0002	0.0003	-0.2369	0.6902	-0.0004	0.0006	-0.0002	0.02	0.00	-2.0607	0.7684
	8	0.0042	0.0021	0.0003	0.0003	0.0005	0.0002	0.7963	0.6873	0.0004	-0.0004	-0.0002	0.0002	-0.0001	-2.8264	0.7769
41	7	1.049	0.0037	0.0018	-5.10	-0.0002	0.0003	-0.3200	0.6333	1.03	0.0001	-0.0002	0.02	1.00	-0.0431	0.6800
	8	0.0037	0.0018	0.0003	0.0003	0.0002	0.0002	0.8264	0.6179	-0.0231	-0.0002	-0.0002	0.0002	-0.0031	0.0452	0.6865
41	7	1.048	0.0039	0.0021	-1.08	-0.0002	0.0002	-0.2813	0.6807	1.04	0.0034	-0.0002	0.02	1.02	-0.3469	0.6978
	8	0.0039	0.0021	0.0002	0.0002	0.0005	0.0002	0.7035	0.5624	-0.0021	-0.0002	-0.0002	0.0002	-0.0003	0.5940	0.8691
41	7	1.049	0.0022	0.0022	-0.04	-0.0002	0.0002	-0.2755	0.6711	1.06	0.0057	-0.0002	0.01	1.03	0.2706	0.7174
	8	0.0022	0.0022	0.0002	0.0002	0.0005	0.0002	0.6525	0.4850	0.0070	-0.0002	-0.0002	0.0002	0.0008	-0.1421	0.5920
41	7	1.047	0.0044	0.0024	0.52	-0.0002	0.0003	-0.2327	0.6790	1.06	0.0116	-0.0001	0.02	1.03	0.1244	0.6881
	8	0.0044	0.0024	0.0003	0.0003	0.0005	0.0002	0.6451	0.5627	0.0112	-0.0001	-0.0001	0.0001	0.0014	-0.0834	0.6325
41	7	1.046	0.0043	0.0021	1.01	-0.0002	0.0003	-0.2325	0.6533	1.07	0.0157	-0.0002	0.02	1.04	0.0799	0.6708
	8	0.0043	0.0021	0.0003	0.0003	0.0005	0.0002	0.7581	0.6401	0.0157	-0.0002	-0.0002	0.0002	0.0019	-0.0716	0.6214
41	7	1.047	0.0036	0.0023	3.12	-0.0001	0.0003	-0.2367	0.6716	1.08	0.0392	-0.0001	0.02	1.05	0.0139	0.6717
	8	0.0036	0.0023	0.0003	0.0003	0.0004	0.0003	0.7306	0.6732	0.0376	-0.0003	-0.0003	0.0003	0.0049	-0.0467	0.6617
41	7	1.045	0.0030	0.0029	6.23	-0.0001	0.0003	-0.2994	0.6851	1.12	0.0717	-0.0001	0.02	1.06	-0.0121	0.6602
	8	0.0030	0.0029	0.0003	0.0003	0.0003	0.0004	0.6127	0.6851	0.0734	-0.0006	-0.0006	0.0006	0.0096	-0.0437	0.6560

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key												
41	8	7	1.049	9.58	-0.0003	-0.0003	-0.3031	0.005	0.005	1.979	0.002	0.002	1.08	-0.0137	0.6428
		8	0.0023	-0.0001	0.0003	0.0003	0.6467	0.7854	0.0008	0.0008	0.1002	-0.0008	0.0125	-0.0424	0.6414
		9	0.0026	0.0003	0.0003	0.0003	0.6657	0.6657	0.0008	0.0008	0.1002	-0.0008	0.0128	-0.0424	0.6414
41	9	7	1.048	12.59	-0.0002	-0.0002	-0.3549	0.05	0.02	1.16	0.02	0.02	1.09	-0.0123	0.6194
		8	0.0020	-0.0001	0.0002	0.0002	1.0384	0.7322	-0.0002	-0.0002	0.1190	-0.0002	0.0147	-0.0362	0.6116
		9	0.0016	0.0003	0.0002	0.0002	0.8140	0.8140	0.0008	0.0008	0.1191	-0.0008	0.0145	-0.0362	0.6116
41	10	7	1.047	-0.002	-0.0003	-0.0003	-0.2587	0.05	0.02	1.06	0.02	0.02	1.02	0.2358	0.6757
		8	0.0041	-0.0002	0.0005	0.0005	0.8494	0.6886	0.0002	0.0002	0.0061	0.0002	0.0007	-0.1706	0.5475
		9	0.0018	0.0003	0.0002	0.0002	0.6252	0.6252	-0.0004	-0.0004	0.0064	-0.0004	0.0007	-0.1706	0.5475
42	1	7	1.045	-3.09	-0.0004	-0.0004	-0.3715	0.05	0.02	3.16	0.02	0.02	3.14	-0.1019	0.6507
		8	0.0034	-0.0002	0.0004	0.0004	0.9806	0.5809	-0.0001	-0.0001	-0.0094	0.0001	-0.0012	0.1688	0.6779
		9	0.0014	0.0002	0.0001	0.0001	0.4590	0.4590	-0.0009	-0.0009	-0.0089	-0.0003	-0.0012	0.1688	0.6779
42	2	7	1.049	-1.02	-0.0005	-0.0005	-0.2811	0.05	0.02	3.18	0.02	0.02	3.16	0.1595	0.7164
		8	0.0038	-0.0002	0.0005	0.0005	0.6831	0.6938	0.0001	0.0001	0.0093	0.0002	0.0013	-0.093E	0.6484
		9	0.0020	0.0002	0.0001	0.0001	0.4756	0.4756	-0.0008	-0.0008	0.0098	-0.0001	0.0012	-0.093E	0.6484
42	3	7	1.050	-0.00	-0.0005	-0.0005	-0.2473	0.05	0.02	3.19	0.02	0.02	3.16	0.0677	0.6888
		8	0.0041	-0.0002	0.0005	0.0005	0.7281	0.6722	0.0002	0.0002	0.0189	0.0002	0.0026	-0.0558	0.6378
		9	0.0021	0.0003	0.0002	0.0002	0.5709	0.5709	-0.0002	-0.0002	0.0187	-0.0002	0.0023	-0.0558	0.6378
42	4	7	1.046	0.55	-0.0005	-0.0005	-0.2272	0.05	0.02	3.19	0.02	0.02	3.17	0.0392	0.6634
		8	0.0043	-0.0001	0.0005	0.0005	0.7517	0.6634	0.0002	0.0002	0.0258	0.0002	0.0031	0.0566	0.6514
		9	0.0021	0.0003	0.0002	0.0002	0.6521	0.6521	-0.0004	-0.0004	0.0243	-0.0004	0.0031	-0.0566	0.6514
42	5	7	1.047	1.03	-0.0005	-0.0005	-0.2406	0.05	0.02	3.20	0.02	0.02	3.17	0.0253	0.6801
		8	0.0040	-0.0001	0.0005	0.0005	0.7693	0.6776	0.0001	0.0001	0.0305	0.0001	0.0041	-0.0534	0.6596
		9	0.0020	0.0003	0.0002	0.0002	0.5933	0.5933	-0.0004	-0.0004	0.0284	-0.0004	0.0037	-0.0534	0.6596
42	6	7	1.049	3.16	-0.0004	-0.0004	-0.2571	0.05	0.02	3.22	0.02	0.02	3.18	-0.0019	0.6744
		8	0.0035	-0.0001	0.0004	0.0004	0.8463	0.6011	-0.0002	-0.0002	0.0542	-0.0002	0.0073	-0.0540	0.6674
		9	0.0019	0.0003	0.0002	0.0002	0.6129	0.6129	0.0005	0.0005	0.0523	-0.0005	0.0069	-0.0540	0.6674
42	7	7	1.048	6.27	-0.0003	-0.0003	-0.2918	0.05	0.02	3.24	0.02	0.02	3.20	-0.0142	0.6544
		8	0.0031	-0.0001	0.0003	0.0003	0.6284	0.5988	-0.0003	-0.0003	0.0836	-0.0003	0.0109	-0.0461	0.6533
		9	0.0028	0.0003	0.0004	0.0004	0.6853	0.6853	-0.0007	-0.0007	0.0853	-0.0007	0.0111	-0.0461	0.6533
42	8	7	1.049	9.39	-0.0004	-0.0004	-0.3155	0.05	0.02	3.27	0.02	0.02	3.21	-0.0096	0.6328
		8	0.0022	-0.0001	0.0003	0.0003	0.5690	0.7394	0.0003	0.0003	0.1069	-0.0003	0.0135	-0.0397	0.6304
		9	0.0030	0.0003	0.0004	0.0004	0.6853	0.6853	-0.0004	-0.0004	0.1097	-0.0004	0.0138	-0.0397	0.6304

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key														
42	9	7	1.048	12.58	-0.0003	-0.0003	-0.2959	0.7604	0.06	3.226	0.1226	-0.0002	0.02	3.24	0.0149	-0.0100	0.6076
		9	0.0021	-0.0001	0.0003	0.0003	0.7961	0.8182	0.1254	0.0009	0.0151	-0.0386	0.6052				
42	10	7	1.048	-0.0000	-0.0000	-0.0000	-0.2591	0.6493	0.05	3.19	0.0194	0.0002	0.02	3.16	0.0026	0.0617	0.6807
		9	0.0041	-0.0002	0.0003	0.0002	0.5370	0.7094	0.0191	0.0002	0.0024	-0.0593	0.6308				
43	3	7	1.048	-3.05	-0.0002	-0.0003	-0.3980	0.6998	0.03	6.20	0.0067	0.0003	0.02	6.24	0.0011	0.2225	0.8416
		9	0.0028	-0.0002	0.0001	0.0001	0.6837	0.4290	0.0089	-0.0002	0.0012	-0.1349	0.7049				
43	4	7	1.047	-1.00	-0.0002	-0.0004	-0.3317	0.7308	0.04	6.21	0.0267	0.0001	0.02	6.26	0.0040	0.0283	0.6997
		9	0.0032	-0.0002	0.0001	0.0001	0.7615	0.4901	0.0292	-0.0002	0.0040	-0.0430	0.6901				
43	5	7	1.047	0.03	-0.0002	-0.0005	-0.2634	0.7000	0.04	6.22	0.0402	0.0000	0.02	6.26	0.0055	0.0060	0.6913
		9	0.0040	-0.0002	0.0002	0.0002	0.6530	0.5085	0.0399	-0.0003	0.0054	-0.0473	0.6806				
43	6	7	1.050	0.56	-0.0002	-0.0005	-0.2475	0.6825	0.04	6.23	0.0456	0.0000	0.02	6.27	0.0062	0.0031	0.6872
		9	0.0041	-0.0002	0.0002	0.0002	0.6562	0.5660	0.0448	-0.0004	0.0060	-0.0484	0.6763				
43	7	7	1.048	1.06	-0.0003	-0.0005	-0.2461	0.7042	0.04	6.23	0.0511	0.0000	0.02	6.27	0.0070	0.0019	0.6880
		9	0.0040	-0.0002	0.0003	0.0002	0.8570	0.6115	0.0512	-0.0005	0.0069	-0.0528	0.6732				
43	8	7	1.050	3.18	-0.0001	-0.0005	-0.2209	0.6889	0.03	6.23	0.0723	0.0001	0.02	6.28	0.0096	0.0120	0.6719
		9	0.0038	-0.0003	0.0003	0.0002	0.8232	0.6598	0.0724	-0.0007	0.0096	-0.0532	0.6662				
43	9	7	1.049	6.30	-0.0001	-0.0003	-0.3206	0.7487	0.04	6.28	0.0985	0.0002	0.02	6.30	0.0127	0.0141	0.6446
		9	0.0024	-0.0003	0.0003	0.0004	0.6711	0.7910	0.1013	-0.0008	0.0129	-0.0436	0.6395				
43	10	7	1.047	8.37	-0.0001	-0.0003	-0.2559	0.7460	0.04	6.30	0.1179	0.0002	0.02	6.31	0.0146	0.0117	0.6205
		9	0.0025	-0.0003	0.0003	0.0004	0.4770	0.7741	0.1185	-0.0008	0.0144	-0.0347	0.5111				
43	11	7	1.048	12.59	-0.0001	-0.0003	-0.3351	0.8204	0.04	6.31	0.1322	0.0003	0.02	6.32	0.0157	0.0128	0.6014
		9	0.0018	-0.0001	0.0003	0.0003	0.5662	0.7279	0.1322	-0.0009	0.0157	-0.0355	0.5950				

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																				
45	1	7	1.048	-5.00	-0.004	-0.5832	0.04	12.25	0.02	12.44	-0.0073	0.7032	0.0030	-0.0002	0.0001	-0.1894	0.7063	0.0464	-0.0000	0.0065	-0.0572	0.6884	
		8	0.0013	0.0003	0.0001	1.1894	0.5353	0.0489	-0.0005	0.0067	-0.0572	0.6884											
45	2	7	1.049	-0.96	-0.005	-0.3005	0.04	12.27	0.02	12.46	-0.0152	0.6819	0.0036	-0.0002	0.0002	0.6447	0.6950	0.0668	-0.0002	0.0091	-0.0557	0.6713	
		8	0.0022	0.0002	0.0002	0.6447	0.6193	0.0692	-0.0007	0.0092	-0.0557	0.6713											
45	3	7	1.047	0.05	-0.005	-0.2528	0.04	12.28	0.02	12.46	-0.0152	0.6704	0.0041	-0.0002	0.0003	0.4901	0.6716	0.0773	-0.0002	0.0103	-0.0521	0.6682	
		8	0.0025	0.0002	0.0003	0.4901	0.6972	0.0800	-0.0008	0.0107	-0.0521	0.6682											
45	4	7	1.048	0.62	-0.005	-0.2626	0.04	12.28	0.02	12.47	-0.0149	0.6641	0.0037	-0.0002	0.0003	0.4901	0.6827	0.0819	-0.0002	0.0108	-0.0514	0.6621	
		8	0.0025	0.0002	0.0003	0.4901	0.6972	0.0848	-0.0008	0.0112	-0.0514	0.6621											
45	5	7	1.051	1.10	-0.005	-0.2235	0.03	12.28	0.02	12.47	-0.0170	0.6595	0.0040	-0.0002	0.0003	0.5946	0.6820	0.0859	-0.0002	0.0113	-0.0520	0.6579	
		8	0.0022	0.0002	0.0003	0.5946	0.7222	0.0876	-0.0009	0.0115	-0.0520	0.6579											
45	6	7	1.046	5.19	-0.004	-0.2269	0.04	12.30	0.02	12.48	-0.0078	0.6373	0.0033	-0.0003	0.0004	0.5738	0.6716	0.1019	-0.0001	0.0129	-0.0457	0.6354	
		8	0.0027	0.0003	0.0004	0.5738	0.7894	0.1054	-0.0009	0.0134	-0.0457	0.6354											
45	7	7	1.049	6.32	-0.004	-0.2351	0.03	12.32	0.02	12.49	-0.0103	0.6156	0.0029	-0.0003	0.0005	0.5302	0.7130	0.1193	-0.0002	0.0144	-0.0311	0.6024	
		8	0.0034	0.0003	0.0005	0.5302	0.7565	0.1200	-0.0007	0.0144	-0.0311	0.6024											
45	8	7	1.050	9.44	-0.003	-0.2363	0.05	12.33	0.03	12.50	-0.0136	0.5944	0.0024	-0.0004	0.0005	0.6513	0.7583	0.1329	-0.0003	0.0159	-0.0336	0.5944	
		8	0.0032	0.0004	0.0005	0.6513	0.8749	0.1351	-0.0009	0.0160	-0.0336	0.5944											
45	9	7	1.048	12.61	0.003	-0.3314	0.05	12.33	0.03	12.50	-0.0225	0.5905	0.0020	-0.0004	0.0003	1.3854	0.7803	0.1396	-0.0006	0.0165	-0.0433	0.5905	
		8	0.0015	0.0004	0.0003	1.3854	1.1019	0.1406	-0.0012	0.0165	-0.0433	0.5905											
45	10	7	1.051	0.05	-0.005	-0.2534	0.03	12.27	0.02	12.46	-0.0170	0.6682	0.0042	-0.0002	0.0003	0.5905	0.6766	0.0769	-0.0002	0.0102	-0.0519	0.6682	
		8	0.0026	0.0003	0.0004	0.5905	0.7809	0.0797	-0.0008	0.0106	-0.0519	0.6682											
46	1	7	1.048	-2.98	-0.004	-0.3547	0.03	14.98	0.03	15.25	-0.0191	0.6904	0.0032	-0.0002	0.0003	1.6351	0.6511	0.0630	-0.0002	0.0087	-0.0615	0.6904	
		8	0.0011	0.0003	0.0001	1.6351	0.5748	0.0675	-0.0008	0.0090	-0.0615	0.6904											

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key														
46	2	7	1.047	-0.95	-0.00	-0.2409	0.04	14.99	0.02	15.26	-0.0172	0.6663				
		8	0.0041	-0.0002	0.0006	0.7213	0.0823	0.0002	0.0109							
		9	0.0027	0.0002	0.0003	0.6948	0.0858	-0.0009	0.0112			0.6560				
46	3	7	1.050	0.08	-0.00	-0.2694	0.03	15.00	0.02	15.26	-0.0157	0.6564				
		8	0.0036	-0.0002	0.0005	0.6742	0.0911	-0.0002	0.0119							
		9	0.0027	0.0002	0.0004	0.7346	0.0955	-0.0009	0.0123			0.6476				
46	4	7	1.051	0.62	-0.00	-0.2455	0.03	15.00	0.02	15.26	-0.0159	0.6508				
		8	0.0038	-0.0001	0.0005	0.6612	0.0953	-0.0003	0.0124							
		9	0.0028	0.0002	0.0003	0.6888	0.0991	-0.0009	0.0127			0.6431				
46	5	7	1.051	1.11	-0.00	-0.2141	0.04	15.00	0.02	15.26	-0.0148	0.6456				
		8	0.0037	-0.0001	0.0005	0.6721	0.0987	-0.0002	0.0127							
		9	0.0025	0.0002	0.0004	0.7808	0.1020	-0.0010	0.0130			0.6383				
46	6	7	1.049	3.23	-0.00	-0.1823	0.05	15.02	0.03	15.27	-0.0135	0.6270				
		8	0.0037	-0.0001	0.0004	0.6619	0.1127	-0.0003	0.0141							
		9	0.0026	0.0004	0.0004	0.7921	0.1128	-0.0007	0.0138			0.6144				
46	7	7	1.048	6.33	-0.00	-0.2159	0.04	15.03	0.03	15.28	-0.0118	0.6073				
		8	0.0027	-0.0001	0.0004	0.7274	0.1268	-0.0003	0.0154							
		9	0.0031	0.0004	0.0005	0.8053	0.1297	-0.0008	0.0154			0.5961				
46	8	7	1.046	9.44	-0.00	-0.2223	0.04	15.04	0.03	15.29	-0.0171	0.5956				
		8	0.0026	-0.0001	0.0003	0.7319	0.1385	-0.0004	0.0164							
		9	0.0031	0.0004	0.0005	0.8121	0.1405	-0.0010	0.0165			0.5882				
46	0	7	1.051	12.61	-0.00	-0.3083	0.04	15.03	0.03	15.28	-0.0280	0.5923				
		8	0.0019	-0.0001	0.0003	0.8613	0.1401	-0.0007	0.0166							
		9	0.0018	0.0004	0.0003	0.8670	0.1425	-0.0014	0.0166			0.5848				
46	10	7	1.047	0.06	-0.00	-0.2299	0.04	15.00	0.02	15.26	-0.0169	0.6535				
		8	0.0041	-0.0001	0.0005	0.7056	0.0914	-0.0003	0.0119							
		9	0.0025	0.0003	0.0004	0.7954	0.0947	-0.0009	0.0122			0.6469				
47	6	7	0.799	-3.12	-0.00	-0.0902	0.04	-3.06	0.02	-3.07	0.1351	0.6544				
		8	0.0035	-0.0000	0.0004	0.7014	-0.0430	-0.0011	-0.0056							
		9	0.0015	0.0003	0.0001	0.3947	-0.0412	-0.0015	-0.0054			0.6582				
47	7	7	0.800	-1.05	-0.00	-0.0536	0.04	-3.03	0.02	-3.05	0.1334	0.6733				
		8	0.0037	-0.0000	0.0005	0.7054	-0.0213	-0.0005	-0.0028							
		9	0.0013	0.0002	0.0000	0.2771	-0.0221	-0.0009	-0.0030			0.6975				

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																
47	8	7	0.799	-0.005	-0.0510	0.04	-3.01	0.02	-3.04	0.1175	0.6719	0.0041	-0.0003	0.0418	-0.0124	-0.0002	-0.0016	0.2476	0.7290
		6	0.0011	0.0001	1.4160	0.5115	-0.0126	-0.0006	-0.0018										
		9	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000										
47	9	7	0.800	-0.005	-0.0514	0.04	-3.00	0.02	-3.02	0.0981	0.7061	0.0041	-0.0003	0.6324	-0.0083	-0.0001	-0.0011	0.2883	0.7779
		6	0.0010	0.0000	1.5166	0.3663	-0.0085	-0.0004	-0.0013										
		9	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000										
47	10	7	0.798	-0.005	0.0123	0.05	-2.99	0.02	-3.02	-0.0475	0.7477	0.0043	-0.0003	0.584	-0.0036	0.0000	-0.0005	0.4204	0.8587
		6	0.0013	0.0003	1.2424	0.3875	-0.0044	-0.0003	-0.0007										
		9	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000										
47	11	7	0.799	-0.004	-0.0096	0.04	-2.97	0.02	-2.99	0.2421	0.6458	0.0038	-0.0003	0.566	0.0138	0.0006	0.0017	0.1018	0.5855
		6	0.0016	0.0003	1.0232	0.3605	0.0124	0.0002	0.0014										
		9	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000										
47	12	7	0.799	-0.003	-0.0889	0.04	-2.92	0.02	-2.95	0.1861	0.6430	0.0026	-0.0003	0.643	0.0427	0.0015	0.0055	0.1339	0.6358
		6	0.0015	0.0003	1.1859	0.4777	0.0429	0.0011	0.0054										
		9	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000										
47	13	7	0.799	-0.002	-0.2268	0.04	-2.88	0.02	-2.91	0.1694	0.5948	0.0022	-0.0001	0.6048	0.0641	0.0021	0.0076	0.1350	0.5838
		6	0.0016	0.0003	1.2003	0.6026	0.0666	0.0018	0.0077										
		9	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000										
47	14	7	0.798	-0.003	-0.1920	0.04	-2.88	0.02	-2.92	0.0784	0.5682	0.0026	-0.0001	0.6213	0.0778	0.0012	0.0088	0.0700	0.5603
		6	0.0013	0.0003	1.2999	0.4627	0.0809	0.0011	0.0090										
		9	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000										
47	15	7	0.800	-0.004	-0.0500	0.04	-3.01	0.02	-3.03	0.1367	0.7033	0.0038	-0.0003	0.6853	-0.0116	-0.0003	-0.0016	0.2436	0.7265
		6	0.0007	0.0003	2.3913	0.4563	-0.0125	-0.0006	-0.0018										
		9	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000										
48	4	7	0.800	-0.004	-0.1115	0.05	-1.04	0.01	-1.04	0.1440	0.6791	0.0032	-0.0003	0.6221	-0.0298	-0.0008	-0.0040	0.1931	0.6734
		6	0.0001	0.0003	8.4042	1.7448	-0.0303	-0.0011	-0.0040										
		9	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000										
48	5	7	0.799	-0.004	-0.0824	0.04	-1.01	0.01	-1.01	0.1218	0.7066	0.0035	-0.0003	0.6829	-0.0100	-0.0002	-0.0014	0.2784	0.7545
		6	0.0004	0.0003	3.8033	1.6370	-0.0091	-0.0005	-0.0013										
		9	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000										
48	6	7	0.800	-0.005	-0.0506	0.04	-0.99	0.02	-1.00	-0.1952	0.8607	0.0037	-0.0003	0.6769	-0.0020	0.0000	-0.0002	3.6109	3.1818
		6	0.0010	0.0003	1.8643	0.9657	-0.0003	-0.0002	-0.0002										
		9	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000										

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																				
48	7	0.799	0.0036	0.0007	0.45	-0.0000	0.0003	-0.0004	0.0001	-0.0594	2.4085	0.04	0.6755	1.2532	-0.0013	0.0027	0.02	0.0002	0.0001	-1.00	0.8644	0.5276
	6	0.0037	0.0007		0.0000																	
48	8	0.799	0.0037	0.0005	0.99	-0.0000	0.0003	-0.0004	0.0001	-0.0428	3.5494	0.04	0.6577	1.4805	-0.0058	0.0058	0.02	0.0004	0.0006	-0.99	0.3509	0.6349
	6	0.0036	0.0006		0.0000																	
48	9	0.800	0.0036	0.0006	3.07	-0.0000	0.0003	-0.0004	0.0001	-0.0257	2.8503	0.04	0.6437	1.3048	-0.0267	0.0240	0.02	0.0010	0.0030	-0.96	0.2012	0.6425
	6	0.0036	0.0006		0.0000																	
48	10	0.798	0.0025	0.0012	6.22	-0.0000	0.0004	-0.0003	0.0002	-0.0900	1.6872	0.04	0.7128	1.1556	-0.0517	0.0528	0.02	0.0019	0.0063	-0.92	0.1858	0.6162
	6	0.0025	0.0012		0.0000																	
48	11	0.796	0.0022	0.0016	9.36	-0.0000	0.0004	-0.0003	0.0003	-0.2148	1.3541	0.04	0.6440	1.0628	-0.0718	0.0731	0.02	0.0021	0.0082	-0.90	0.1495	0.5809
	6	0.0022	0.0016		0.0000																	
48	12	0.800	0.0021	0.0007	12.51	-0.0000	0.0003	-0.0002	0.0001	-0.2457	2.6173	0.04	0.6182	1.3670	-0.0804	0.0825	0.02	0.0009	0.0092	-0.91	0.0593	0.5628
	6	0.0021	0.0007		0.0000																	
48	13	0.799	0.0042	0.0006	-0.0003	-0.0000	0.0004	-0.0005	0.0002	-0.0398	3.4522	0.04	0.6351	1.6881	-0.0017	0.0004	0.02	0.0002	-0.0002	-1.00	-0.1629	1.0390
	6	0.0042	0.0006		0.0000																	
49	1	0.781	0.0036	0.0003	-4.12	-0.0000	0.0003	-0.0004	0.0001	-0.1182	5.0052	0.05	0.6188	1.6469	-0.0241	0.0244	0.01	0.0007	-0.0032	-0.03	0.1531	0.6757
	6	0.0036	0.0003		0.0000																	
49	0	0.799	0.0038	0.0004	-1.06	-0.0000	0.0003	-0.0005	0.0001	-0.0677	4.2848	0.04	0.6603	1.6943	-0.0050	0.0047	0.01	0.0001	-0.0008	-0.00	0.1050	0.7980
	6	0.0038	0.0004		0.0000																	
49	3	0.799	0.0038	0.0005	-0.0004	-0.0000	0.0003	-0.0005	0.0001	-0.0633	3.3275	0.05	0.6611	1.4466	0.0023	0.0035	0.01	0.0002	0.0002	-0.00	0.5005	0.5463
	6	0.0038	0.0005		0.0000																	
49	4	0.799	0.0043	0.0007	0.45	-0.0000	0.0004	-0.0005	0.0002	-0.0223	2.8574	0.04	0.6216	1.4953	0.0062	0.0058	0.01	0.0003	0.0006	0.00	0.3023	0.6194
	6	0.0043	0.0007		0.0000																	

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key															
49	5	7	0.798	1.01	-0.004	-0.0004	0.0004	0.0001	-0.0004	0.0406	0.6075	0.0112	0.0005	0.01	0.0014	0.2413	0.6430	
		8	0.0039	-0.0004	0.0004	0.0001	0.0004	0.0001	-0.0406	1.1365	0.0104	0.0001	0.0005	0.0112	0.0012	0.0820	0.5822	
		9	0.0007	0.0004	0.0004	0.0001	0.0004	0.0001	2.6076	1.7138	0.0303	0.0007	0.02	0.04	0.0039	0.1796	0.6470	
49	6	7	0.798	3.09	-0.005	0.0002	0.0002	-0.0005	-0.0035	0.6472	0.0303	0.0011	0.02	0.03	0.1248	0.6432		
		8	0.0040	-0.0004	0.0002	0.0002	0.0002	0.0005	3.3937	1.7138	0.0303	0.0007	0.02	0.04	0.0039	0.1796	0.6470	
		9	0.0006	0.0004	0.0002	0.0002	0.0002	0.0005	5.3937	1.7138	0.0303	0.0007	0.02	0.04	0.0039	0.1248	0.6432	
49	7	7	0.796	6.23	-0.003	0.0002	0.0002	-0.0003	-0.1178	0.6301	0.0565	0.0020	0.01	0.08	0.1806	0.5992		
		8	0.0025	-0.0004	0.0002	0.0002	0.0002	0.0003	2.6882	1.5120	0.0565	0.0016	0.0020	0.01	0.067	0.1479	0.6020	
		9	0.0008	0.0004	0.0002	0.0002	0.0002	0.0003	2.6882	1.5120	0.0565	0.0016	0.0020	0.01	0.067	0.1479	0.6020	
49	8	7	0.799	9.34	-0.003	0.0002	0.0002	-0.0003	-0.1538	0.6703	0.0735	0.0018	0.01	0.09	0.1229	0.5687		
		8	0.0024	-0.0004	0.0002	0.0002	0.0002	0.0003	1.8186	1.1866	0.0735	0.0018	0.01	0.09	0.1002	0.5655		
		9	0.0012	0.0004	0.0002	0.0002	0.0002	0.0003	1.8186	1.1866	0.0735	0.0018	0.01	0.09	0.1002	0.5655		
49	9	7	0.797	12.50	-0.002	0.0002	0.0002	-0.0003	-0.2457	0.6182	0.0823	0.0008	0.02	0.07	0.0534	0.5638		
		8	0.0022	-0.0004	0.0002	0.0002	0.0002	0.0003	2.6424	1.3762	0.0823	0.0005	0.0008	0.02	0.0093	0.0352	0.5563	
		9	0.0008	0.0004	0.0002	0.0002	0.0002	0.0003	2.6424	1.3762	0.0823	0.0005	0.0008	0.02	0.0093	0.0352	0.5563	
49	10	7	0.798	-0.003	-0.005	0.0002	0.0002	-0.0003	-0.0258	0.6207	0.0038	0.0002	0.01	0.00	0.4275	0.5234		
		8	0.0043	-0.0004	0.0002	0.0002	0.0002	0.0003	2.2250	1.1234	0.0038	0.0002	0.01	0.00	-0.1476	0.3890		
		9	0.0009	0.0004	0.0002	0.0002	0.0002	0.0003	2.2250	1.1234	0.0038	0.0002	0.01	0.00	-0.1476	0.3890		
50	1	7	0.798	-3.10	-0.004	0.0000	0.0000	-0.0004	-0.1182	0.6188	-0.0172	-0.0005	0.01	0.96	0.1638	0.6966		
		8	0.0034	-0.0000	0.0000	0.0000	0.0000	0.0004	15.9065	28.9656	-0.0172	-0.0008	-0.0005	0.01	0.023	0.2336	0.6924	
		9	0.0000	0.0003	0.0000	0.0000	0.0000	0.0004	15.9065	28.9656	-0.0172	-0.0008	-0.0005	0.01	0.023	0.2336	0.6924	
50	2	7	0.798	-1.03	-0.004	0.0001	0.0001	-0.0004	-0.0811	0.6578	1.00	0.001	0.01	0.99	-6.9022	5.9755		
		8	0.0037	-0.0000	0.0001	0.0001	0.0001	0.0004	4.0166	1.6941	1.00	0.001	0.01	0.99	-0.8031	0.0358		
		9	0.0005	0.0004	0.0001	0.0001	0.0001	0.0004	4.0166	1.6941	1.00	0.001	0.01	0.99	-0.8031	0.0358		
50	3	7	0.798	-0.02	-0.005	0.0001	0.0001	-0.0004	-0.0634	0.6730	1.02	0.004	0.01	1.00	0.2641	0.6438		
		8	0.0037	-0.0000	0.0001	0.0001	0.0001	0.0004	4.7259	1.9147	1.02	0.004	0.004	1.00	0.0673	0.5826		
		9	0.0004	0.0004	0.0001	0.0001	0.0001	0.0004	4.7259	1.9147	1.02	0.004	0.004	1.00	0.0673	0.5826		
50	4	7	0.798	0.46	-0.004	0.0001	0.0001	-0.0004	-0.0604	0.6018	1.03	0.005	0.01	1.01	0.2242	0.6507		
		8	0.0040	-0.0000	0.0001	0.0001	0.0001	0.0004	3.3035	1.4480	1.03	0.005	0.005	1.01	0.0971	0.6066		
		9	0.0006	0.0004	0.0001	0.0001	0.0001	0.0004	3.3035	1.4480	1.03	0.005	0.005	1.01	0.0971	0.6066		
50	5	7	0.798	0.99	-0.005	0.0001	0.0001	-0.0004	-0.0218	0.6220	1.04	0.007	0.01	1.01	0.1987	0.6434		
		8	0.0042	-0.0000	0.0001	0.0001	0.0001	0.0004	2.8552	1.3209	1.04	0.007	0.007	1.01	0.1087	0.6277		
		9	0.0007	0.0004	0.0001	0.0001	0.0001	0.0004	2.8552	1.3209	1.04	0.007	0.007	1.01	0.1087	0.6277		

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																					
50	6	7	0.799	5.06	-0.004	-0.0480	0.6386	0.05	1.08	0.01	1.04	0.1734	0.6457	0.0034	-0.0000	0.0004	0.0001	3.4381	0.6386	0.0391	0.0013	0.0050	0.1231	0.6499
		8	0.0006	0.0004	0.0001	3.4381	1.4788	0.0385	0.0385	0.0009	0.0050	0.1231	0.6457	0.0034	0.0000	0.0004	0.0001	3.4381	0.6386	0.0391	0.0013	0.0050	0.1231	0.6499
50	7	8	0.798	6.22	-0.003	-0.1197	0.6226	0.05	1.11	0.01	1.08	0.1701	0.5957	0.0024	-0.0000	0.0002	0.0002	2.1535	0.6226	0.0603	0.0020	0.0071	0.1394	0.5892
		9	0.0010	0.0004	0.0002	2.1535	1.1702	0.0605	0.0605	0.0016	0.0071	0.1394	0.5892	0.0024	0.0000	0.0002	0.0002	2.1535	0.6226	0.0603	0.0020	0.0071	0.1394	0.5892
50	8	7	0.797	9.35	-0.002	-0.2546	0.6372	0.04	1.11	0.01	1.08	0.0982	0.5664	0.0021	-0.0001	0.0002	0.0002	-2.3124	0.6372	0.0781	0.0014	0.0085	0.0982	0.5664
		8	0.0009	0.0004	0.0002	2.3124	1.4057	0.0781	0.0781	0.0013	0.0085	0.0982	0.5664	0.0009	0.0000	0.0002	0.0002	2.3124	0.6372	0.0781	0.0014	0.0085	0.0982	0.5664
50	9	7	0.797	12.53	-0.002	-0.2568	0.5966	0.04	1.11	0.01	1.07	0.0456	0.5621	0.0022	-0.0001	0.0002	0.0002	-2.7622	0.5966	0.0832	0.0007	0.0093	0.0456	0.5621
		8	0.0008	0.0004	0.0002	2.7622	1.4686	0.0821	0.0821	0.0004	0.0093	0.0456	0.5621	0.0008	0.0000	0.0002	0.0002	2.7622	0.5966	0.0821	0.0004	0.0093	0.0456	0.5621
50	10	7	0.799	-0.00	-0.005	-0.0175	0.6411	0.05	1.02	0.01	1.00	0.2418	0.6094	0.0042	-0.0000	0.0004	0.0004	3.7549	0.6411	0.0084	0.0004	0.0010	0.2418	0.6094
		8	0.0006	0.0004	0.0002	3.7549	1.7196	0.0084	0.0084	0.0000	0.0010	0.2418	0.6094	0.0006	0.0000	0.0004	0.0004	3.7549	0.6411	0.0084	0.0004	0.0010	0.2418	0.6094
51	3	7	0.799	-3.08	-0.004	-0.0886	0.7223	0.05	2.99	0.01	2.98	0.0682	0.6409	0.0032	-0.0000	0.0004	0.0004	1.7408	0.7223	0.0065	0.0000	0.0008	0.0682	0.6409
		8	0.0007	0.0002	0.0000	1.7408	0.6324	0.0065	0.0065	0.0004	0.0008	0.0682	0.6409	0.0007	0.0000	0.0004	0.0004	1.7408	0.7223	0.0065	0.0000	0.0008	0.0682	0.6409
51	4	7	0.800	-1.02	-0.005	-0.0542	0.6711	0.05	3.01	0.01	3.00	0.3001	0.7336	0.0039	-0.0000	0.0003	0.0003	1.1253	0.6711	0.0091	0.0005	0.0012	0.3001	0.7336
		8	0.0013	0.0003	0.0001	1.1253	0.6984	0.0101	0.0101	0.0001	0.0012	0.3001	0.7336	0.0013	0.0000	0.0003	0.0003	1.1253	0.6984	0.0091	0.0005	0.0012	0.3001	0.7336
51	5	7	0.799	0.01	-0.006	0.0074	0.7619	0.05	3.03	0.02	3.01	0.2180	0.6852	0.0041	0.0000	0.0002	0.0001	0.0715	0.7619	0.0192	0.0008	0.0023	0.2180	0.6852
		8	0.0013	0.0002	0.0001	0.0715	0.5580	0.0179	0.0179	0.0003	0.0023	0.2180	0.6852	0.0013	0.0000	0.0002	0.0001	0.0715	0.5580	0.0179	0.0003	0.0023	0.2180	0.6852
51	6	7	0.799	0.48	-0.005	-0.0248	0.7010	0.05	3.05	0.02	3.02	0.2008	0.6791	0.0038	-0.0000	0.0003	0.0001	1.9107	0.7010	0.0245	0.0009	0.0033	0.2008	0.6791
		8	0.0008	0.0003	0.0001	1.9107	0.9582	0.0235	0.0235	0.0005	0.0033	0.2008	0.6791	0.0008	0.0000	0.0003	0.0001	1.9107	0.9582	0.0235	0.0005	0.0033	0.2008	0.6791
51	7	7	0.799	1.00	-0.005	-0.0123	0.6867	0.05	3.06	0.02	3.02	0.1913	0.6740	0.0037	-0.0000	0.0003	0.0001	0.5454	0.6867	0.0301	0.0011	0.0040	0.1913	0.6740
		8	0.0010	0.0003	0.0001	0.5454	0.7952	0.0284	0.0284	0.0006	0.0040	0.1913	0.6740	0.0010	0.0000	0.0003	0.0001	0.5454	0.7952	0.0284	0.0006	0.0040	0.1913	0.6740
51	8	7	0.799	3.16	-0.005	0.0239	0.7295	0.05	3.09	0.02	3.05	0.1961	0.6358	0.0037	0.0000	0.0002	0.0002	1.0776	0.7295	0.0491	0.0018	0.0061	0.1961	0.6358
		8	0.0015	0.0003	0.0002	1.0776	0.7552	0.0490	0.0490	0.0012	0.0061	0.1961	0.6358	0.0015	0.0000	0.0002	0.0002	1.0776	0.7552	0.0490	0.0012	0.0061	0.1961	0.6358

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key															
51	9	0.799	0.0024	-0.0000	6.30	0.0000	-0.0003	0.0003	-0.0003	-0.0708	0.710	0.05	3.11	0.020	3.08	0.1500	0.5949
	8	0.0013	0.0003	0.0003	0.0003	0.0002	0.0002	0.0002	0.0002	1.2471	0.8312	0.0704	0.0016	0.0082	0.0081	0.1141	0.5841
	9	0.797	0.0021	-0.0000	9.37	0.0000	-0.0003	0.0003	-0.0003	-0.2019	0.7028	0.04	3.10	0.02	3.07	0.0714	0.5684
51	10	0.0020	0.0003	0.0003	0.0003	0.0003	0.0003	0.0003	0.0003	0.9415	0.8175	0.0813	0.0008	0.0091	0.0091	0.0544	0.5612
	7	0.799	0.0023	-0.0000	12.52	0.0000	0.0003	0.0003	0.0003	-0.1255	0.8326	0.05	3.10	0.01	3.07	0.0427	0.5682
	8	0.0015	0.0003	0.0003	0.0003	0.0002	0.0002	0.0002	0.0002	1.1743	0.8710	0.0839	0.0004	0.0094	0.0094	0.0251	0.5602
51	12	0.799	0.0040	-0.0000	-0.01	0.0003	-0.0005	0.0005	-0.0005	-0.0261	0.6869	0.05	3.04	0.01	3.02	0.2152	0.6818
	8	0.0010	0.0003	0.0003	0.0003	0.0001	0.0001	0.0001	0.0001	1.5848	0.7291	0.0189	0.0004	0.0024	0.0024	0.1100	0.6445
	9	0.800	0.0032	-0.0000	-3.05	0.0003	-0.0004	0.0004	-0.0004	-0.0886	0.7223	0.04	6.00	0.02	6.00	0.2856	0.7330
52	1	0.800	0.0011	0.0003	0.0003	0.0003	0.0003	0.0003	0.0003	2.3210	0.6064	0.0085	0.0000	0.0010	0.0010	0.0518	0.6365
	7	0.800	0.0040	-0.0000	-0.98	0.0003	-0.0005	0.0005	-0.0005	-0.0300	0.6863	0.03	6.03	0.01	6.04	0.1819	0.6833
	8	0.0011	0.0003	0.0003	0.0003	0.0001	0.0001	0.0001	0.0001	1.6084	0.7738	0.0279	0.0005	0.0036	0.0036	0.1069	0.6583
52	3	0.800	0.0032	-0.0000	0.01	0.0003	-0.0004	0.0004	-0.0004	-0.0594	0.7316	0.05	6.05	0.02	6.05	0.1732	0.6683
	8	0.0011	0.0003	0.0003	0.0003	0.0001	0.0001	0.0001	0.0001	1.4639	0.6182	0.0382	0.0008	0.0051	0.0051	0.1148	0.6613
	9	0.799	0.0036	-0.0000	0.51	0.0003	-0.0005	0.0005	-0.0005	-0.0053	0.7422	0.04	6.05	0.02	6.05	0.1748	0.6610
52	4	0.800	0.0009	0.0003	0.0003	0.0003	0.0003	0.0003	0.0003	1.7926	0.8703	0.0417	0.0009	0.0054	0.0054	0.1139	0.6541
	7	0.800	0.0039	0.0000	1.05	0.0003	-0.0005	0.0005	-0.0005	0.0113	0.7100	0.04	6.07	0.02	6.06	0.1749	0.6440
	8	0.0013	0.0003	0.0003	0.0003	0.0001	0.0001	0.0001	0.0001	1.3167	0.6637	0.0470	0.0011	0.0060	0.0060	0.1210	0.6398
52	6	0.800	0.0033	0.0000	3.19	0.0003	-0.0004	0.0004	-0.0004	0.0172	0.7394	0.0611	0.0019	0.0073	0.0073	0.1613	0.6022
	8	0.0010	0.0003	0.0003	0.0003	0.0001	0.0001	0.0001	0.0001	1.7435	0.8994	0.0595	0.0016	0.0071	0.0071	0.1355	0.6008
	9	0.799	0.0024	-0.0000	6.28	0.0003	-0.0003	0.0003	-0.0003	-0.0708	0.7710	0.04	6.09	0.02	6.08	0.0957	0.5714
52	7	0.0015	0.0003	0.0003	0.0003	0.0002	0.0002	0.0002	0.0002	1.2603	0.8195	0.0771	0.0013	0.0087	0.0087	0.0867	0.5636
	8	0.799	0.0024	-0.0000	6.28	0.0003	-0.0003	0.0003	-0.0003	-0.0708	0.7710	0.04	6.09	0.02	6.08	0.0957	0.5714
	9	0.0015	0.0003	0.0003	0.0003	0.0002	0.0002	0.0002	0.0002	1.2603	0.8195	0.0771	0.0013	0.0087	0.0087	0.0867	0.5636

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd																					
52	8	7	0.798	9.34	-0.003	-0.003	-0.003	-0.003	0.04	6.08	0.02	6.07	0.0497	0.5669									
	8	8	0.0023	-0.0000	0.0003	0.0003	0.0003	0.8049	0.8444	0.0821	0.0008	0.0093	0.0286	0.5586									
	9	9	0.0021	0.0004	0.0003	0.0003	0.8444	0.9719	0.8444	0.0818	0.0004	0.0091											
52	9	7	0.798	12.53	0.003	0.001	0.04	0.2411	0.373	6.08	0.02	6.07	0.0359	0.5657									
	8	8	0.0020	-0.0000	0.0003	0.0001	0.3841	2.2746	1.3841	0.0866	0.0006	0.0098	0.0152	0.5570									
	9	9	0.0006	0.0003	0.0001	0.0001	1.3841	2.2746	0.8859	0.0859	0.0002	0.0095											
52	10	7	0.799	0.01	-0.005	-0.005	0.04	-0.0415	0.6908	6.05	0.02	6.04	0.1697	0.6648									
	8	8	0.0037	-0.0000	0.0005	0.0001	0.6908	-0.0415	0.6908	0.378	0.0012	0.0050	0.1102	0.6595									
	9	9	0.0008	0.0003	0.0001	0.0001	0.7894	2.0153	0.7894	0.0367	0.0008	0.0048											
53	1	7	0.799	-5.03	-0.004	-0.001	0.04	-0.0894	0.6985	9.04	0.02	9.01	0.1810	0.7004									
	8	8	0.0033	-0.0000	0.0004	0.0001	0.6985	-0.0894	0.6985	0.260	0.0009	0.0036	0.0998	0.6670									
	9	9	0.0009	0.0003	0.0001	0.0001	0.5728	1.7335	0.5728	0.0269	0.0005	0.0035											
53	2	7	0.799	-0.97	-0.005	-0.001	0.04	-0.0539	0.7249	9.08	0.02	9.04	0.1673	0.6569									
	8	8	0.0035	-0.0000	0.0005	0.0001	0.7249	-0.0539	0.7249	0.453	0.0015	0.0059	0.1160	0.6442									
	9	9	0.0011	0.0003	0.0001	0.0001	0.5840	1.5558	0.5840	0.0463	0.0010	0.0059											
53	3	7	0.799	0.04	-0.005	-0.001	0.04	-0.0115	0.7074	9.09	0.02	9.05	0.1780	0.6343									
	8	8	0.0037	-0.0000	0.0005	0.0001	0.7074	-0.0115	0.7074	0.517	0.0018	0.0065	0.1258	0.6189									
	9	9	0.0011	0.0003	0.0001	0.0001	0.5840	1.5558	0.5840	0.0526	0.0013	0.0065											
53	4	7	0.799	0.52	-0.005	-0.001	0.04	-0.0167	0.6942	9.09	0.02	9.05	0.1719	0.6173									
	8	8	0.0038	-0.0000	0.0005	0.0001	0.6942	-0.0167	0.6942	0.551	0.0018	0.0068	0.1331	0.6144									
	9	9	0.0009	0.0003	0.0001	0.0001	0.6693	1.9195	0.6693	0.0559	0.0014	0.0068											
53	5	7	0.800	1.05	-0.005	-0.001	0.04	-0.0307	0.7334	9.10	0.02	9.06	0.1607	0.6099									
	8	8	0.0034	-0.0000	0.0005	0.0001	0.7334	-0.0307	0.7334	0.596	0.0019	0.0072	0.1289	0.6010									
	9	9	0.0006	0.0003	0.0001	0.0001	0.8138	2.5941	0.8138	0.0597	0.0015	0.0071											
53	6	7	0.799	5.17	-0.004	-0.001	0.04	-0.0220	0.6846	9.11	0.01	9.07	0.1301	0.5872									
	8	8	0.0032	-0.0000	0.0004	0.0001	0.6846	-0.0220	0.6846	0.736	0.0019	0.0086	0.0995	0.5829									
	9	9	0.0011	0.0003	0.0001	0.0001	0.8176	1.7104	0.8176	0.0735	0.0014	0.0085											
53	7	7	0.800	6.27	-0.003	-0.001	0.04	-0.0700	0.6965	9.11	0.02	9.06	0.0600	0.5684									
	8	8	0.0027	-0.0000	0.0003	0.0001	0.6965	-0.0700	0.6965	0.801	0.0009	0.0091	0.0424	0.5596									
	9	9	0.0012	0.0004	0.0002	0.0001	0.9358	1.6897	0.9358	0.0811	0.0006	0.0090											
53	8	7	0.799	9.37	-0.003	-0.001	0.04	-0.2146	0.8323	9.11	0.02	9.06	0.0384	0.5670									
	8	8	0.0019	-0.0000	0.0003	0.0001	0.8323	-0.2146	0.8323	0.866	0.0006	0.0097	0.0196	0.5593									
	9	9	0.0020	0.0004	0.0003	0.0001	0.8072	1.0590	0.8072	0.0864	0.0003	0.0096											

See Key

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key														
53	9	0.800	12.54	0.003	-0.1934	0.04	9.11	0.02	9.05	0.0267	0.5655					
	8	0.0022	-0.0000	0.0002	1.5937	0.7680	0.0893	0.0004	0.0101	0.0064	0.5603					
	9	0.0013	0.0004	0.0000	-0.0002	1.0372	0.0891	0.0001	0.0099							
53	10	0.799	0.02	-0.005	-0.0215	0.04	9.09	0.01	9.05	0.1723	0.6258					
	8	0.0040	-0.0000	0.0005	2.2155	0.6992	0.521	0.0017	0.0065	0.1309	0.6269					
	9	0.0008	0.0003	0.0001	-0.0001	0.7509	0.523	0.0013	0.0065							
53	11	0.800	-2.98	-0.00	-0.0929	0.05	12.07	0.01	12.06	0.1556	0.6572					
	8	0.0033	-0.0000	0.0004	2.8572	0.7069	0.436	0.0013	0.0059	0.1087	0.6504					
	9	0.0006	0.0003	0.0000	-0.0000	0.3596	0.454	0.0009	0.0059							
53	12	0.799	-0.98	-0.00	-0.0785	0.04	12.10	0.01	12.09	0.1566	0.6139					
	8	0.0034	-0.0000	0.0004	2.2926	0.6857	0.576	0.0018	0.0070	0.1293	0.6090					
	9	0.0008	0.0003	0.0000	-0.0004	0.3793	0.593	0.0015	0.0072							
53	13	0.799	0.06	-0.005	-0.0252	0.04	12.11	0.01	12.09	0.1359	0.6021					
	8	0.0038	-0.0000	0.0005	1.5743	0.6909	0.655	0.0017	0.0078	0.1071	0.5958					
	9	0.0011	0.0003	0.0001	-0.0001	0.5552	0.657	0.0014	0.0078							
54	1	0.799	-3.01	-0.00	-0.1112	0.04	12.08	0.01	12.06	0.1573	0.6566					
	8	0.0031	-0.0000	0.0004	3.2554	0.6961	0.437	0.0013	0.0059	0.1088	0.6505					
	9	0.0005	0.0003	0.0000	-0.0004	0.2203	0.457	0.0009	0.0059							
54	2	0.799	-0.96	-0.005	-0.0487	0.04	12.10	0.02	12.09	0.1573	0.6125					
	8	0.0036	-0.0000	0.0005	2.0894	0.7329	0.575	0.0018	0.0070	0.1289	0.6090					
	9	0.0008	0.0003	0.0000	-0.0000	0.4649	0.588	0.0015	0.0071							
54	3	0.800	0.06	-0.004	-0.0405	0.04	12.10	0.01	12.09	0.1371	0.6012					
	8	0.0034	-0.0000	0.0004	1.9762	0.7180	0.655	0.0017	0.0077	0.1073	0.5945					
	9	0.0009	0.0003	0.0000	-0.0000	0.4544	0.654	0.0014	0.0077							
54	4	0.800	0.54	-0.004	-0.0419	0.04	12.12	0.01	12.10	0.1313	0.5972					
	8	0.0036	-0.0000	0.0004	2.4558	0.6802	0.692	0.0018	0.0082	0.1007	0.5947					
	9	0.0007	0.0003	0.0000	-0.0000	0.5049	0.694	0.0013	0.0082							
54	5	0.800	1.05	-0.004	-0.0328	0.04	12.12	0.02	12.09	0.1272	0.5914					
	8	0.0035	-0.0000	0.0004	1.8452	0.6857	0.722	0.0018	0.0085	0.0957	0.5888					
	9	0.0010	0.0003	0.0000	-0.0000	0.4094	0.727	0.0013	0.0085							
54	6	0.800	3.18	-0.004	-0.0368	0.04	12.11	0.01	12.10	0.0808	0.5670					
	8	0.0031	-0.0000	0.0004	1.5310	0.6823	0.783	0.0012	0.0088	0.0715	0.5609					
	9	0.0012	0.0003	0.0001	-0.0001	0.4513	0.792	0.0011	0.0090							

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd																		
54	7	0.799	6.25	-0.003	0.04	12.10	0.01	12.08	0.0440	0.5650									
	6	0.0024	-0.0000	0.0001	0.8102	0.0833	0.0007	0.0094	0.0246	0.5597									
	9	0.0011	0.0004	0.0001	0.5959	0.0829	0.0004	0.0092											
54	7	0.800	9.38	-0.003	0.04	12.11	0.01	12.09	0.0314	0.5652									
	8	0.0020	-0.0000	0.0003	0.7910	0.0888	0.0005	0.0100	0.0127	0.5610									
	9	0.0013	0.0003	0.0001	0.7295	0.0880	0.0002	0.0098											
54	7	0.800	12.54	0.003	0.04	12.11	0.01	12.09	0.0186	0.5637									
	6	0.0023	-0.0000	0.0003	0.7679	0.0937	0.0003	0.0105	-0.0000	0.5632									
	9	0.0015	0.0004	0.0002	0.7739	0.0941	-0.0000	0.0106											
54	7	0.800	0.06	-0.003	0.04	12.10	0.01	12.09	0.1347	0.5985									
	8	0.0037	-0.0000	0.0005	0.7275	0.0659	0.0017	0.0079	0.1070	0.5941									
	9	0.0009	0.0003	0.0000	0.4365	0.0663	0.0014	0.0078											
55	7	0.800	-2.99	-0.003	0.04	15.09	0.02	15.09	0.1563	0.6215									
	6	0.0029	-0.0000	0.0003	0.6425	0.0560	0.0017	0.0069	0.1250	0.6110									
	9	0.0004	0.0003	-0.0000	0.2268	0.0591	0.0014	0.0072											
55	7	0.799	-0.95	-0.005	0.04	15.11	0.02	15.09	0.1271	0.6008									
	6	0.0036	-0.0000	0.0005	0.7374	0.0709	0.0016	0.0085	0.0968	0.5903									
	9	0.0008	0.0003	0.0000	0.3684	0.0722	0.0013	0.0085											
55	7	0.800	0.09	-0.005	0.04	15.12	0.01	15.10	0.1172	0.5854									
	6	0.0035	-0.0000	0.0005	0.7296	0.0760	0.0017	0.0089	0.0860	0.5739									
	9	0.0009	0.0003	0.0000	0.3140	0.0765	0.0013	0.0087											
55	7	0.800	0.55	-0.004	0.04	15.12	0.01	15.09	0.1050	0.5783									
	6	0.0034	-0.0000	0.0004	0.7172	0.0764	0.0016	0.0088	0.0796	0.5692									
	9	0.0012	0.0003	0.0000	0.2920	0.0774	0.0012	0.0088											
55	7	0.800	1.12	-0.005	0.04	15.11	0.02	15.09	0.0909	0.5746									
	6	0.0034	0.0000	0.0005	0.7584	0.0775	0.0014	0.0089	0.0716	0.5665									
	9	0.0013	0.0003	0.0000	0.3581	0.0786	0.0011	0.0089											
55	7	0.800	3.17	-0.004	0.04	15.11	0.01	15.09	0.0574	0.5704									
	6	0.0033	-0.0000	0.0004	0.6941	0.0811	0.0009	0.0092	0.0377	0.5602									
	9	0.0011	0.0003	0.0001	0.4899	0.0821	0.0006	0.0092											
55	7	0.801	6.28	-0.002	0.04	15.10	0.01	15.09	0.0350	0.5683									
	6	0.0019	-0.0000	0.0002	0.7192	0.0875	0.0006	0.0099	0.0168	0.5624									
	9	0.0007	0.0004	0.0000	0.6426	0.0875	0.0002	0.0098											

See Key

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key															
55	8	0.801	0.0023	0.0004	0.0003	-0.0990	0.05	15.10	0.02	15.08	0.0232	0.5667					
	9	0.0021	-0.0004	0.0004	0.0002	1.0401	0.7880	0.0908	0.0004	0.0102	0.0044	0.5625					
55	0	0.800	0.0021	0.0004	0.0003	-0.2227	0.04	15.11	0.01	15.08	0.0148	0.5644					
	5	0.0018	-0.0004	0.0004	0.0002	1.1427	0.7139	0.0982	0.0002	0.0110	-0.0057	0.5595					
55	10	0.801	0.0039	0.0003	0.0005	-0.0174	0.05	15.12	0.01	15.10	0.1162	0.5846					
	6	0.0012	-0.0003	0.0003	0.0001	1.5859	0.6920	0.0759	0.0017	0.0088	0.0853	0.5763					
56	3	0.799	0.0049	0.0001	0.0006	-0.0065	0.53	-0.53	-0.48	0.46	0.1403	0.6689					
	8	0.0011	-0.0001	0.0001	-0.0002	-0.7863	0.6869	-0.266	-0.0007	-0.0035	0.2095	0.6592					
56	4	0.799	0.0044	0.0001	0.0006	-0.0077	0.53	-0.54	-0.48	0.46	0.1437	0.6733					
	9	0.0014	-0.0001	0.0001	-0.0002	-0.6398	0.7303	-0.265	-0.0007	-0.0028	0.2115	0.6622					
56	5	0.799	0.0051	0.0001	0.0007	0.0270	0.53	-0.53	-0.48	0.46	0.1435	0.6741					
	9	0.0011	0.0001	0.0001	-0.0002	-0.7861	0.7236	-0.268	-0.0007	-0.0029	0.2087	0.6598					
56	6	0.799	0.0061	0.0002	0.0008	0.0665	0.54	-0.48	-0.48	0.50	-7.8661	0.2929					
	5	0.0005	0.0002	0.0002	-0.0000	-2.1495	0.9123	-0.0037	-0.0000	-0.0004	0.0602	0.6270					
56	7	0.799	0.0060	0.0002	0.0008	0.2468	0.53	-0.48	-0.48	0.51	0.5014	0.7129					
	9	0.0007	0.0002	0.0002	-0.0001	-1.5136	0.6646	0.0037	0.0003	0.0010	0.0797	0.6549					
56	8	0.783	0.0059	0.0002	0.0007	0.0445	0.53	-0.47	-0.48	0.51	0.2934	0.6656					
	9	0.0007	0.0002	0.0002	-0.0001	-1.6265	0.8781	0.0137	0.0003	0.0017	0.1156	0.6435					
56	9	0.798	0.0057	0.0002	0.0007	0.0529	0.53	-0.45	-0.48	0.53	0.2192	0.6568					
	9	0.0009	0.0002	0.0002	-0.0000	-1.2151	0.6478	0.0190	0.0008	0.0024	0.1214	0.6521					
56	10	0.799	0.0057	0.0002	0.0007	0.0471	0.53	-0.43	-0.48	0.54	0.1928	0.6593					
	9	0.0009	0.0002	0.0002	-0.0001	-1.2191	0.6194	0.0302	-0.0011	0.0039	0.1240	0.6516					

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																					
56	11	7	0.798	4.16	-0.00	0.0490	0.53	-0.41	-0.48	0.55	0.1048	0.6483	7	0.799	-0.03	0.0254	0.53	-0.49	-0.48	0.50	7.2388	-0.5234	
		6	0.0048	0.0000	0.0006	0.0490	0.6806	0.0396	0.0014	0.0051	0.1283	0.6409		6	0.0057	0.0000	0.6465	0.0001	0.0001	-0.0000	0.0000	-0.0634	0.4872
		5	-0.0008	0.0002	-0.0001	-1.4381	0.6821	0.0429	0.0011	0.0055				5	-0.0009	0.0002	0.9032	0.0044	-0.0000	0.0004			
57	1	7	0.799	-5.10	-0.00	0.0587	1.02	-1.05	-0.99	0.95	0.1439	0.6690	7	0.799	-5.10	0.0009	0.6902	-1.05	-0.99	0.95	0.1439	0.6690	
		6	0.0071	0.0000	0.0009	0.0587	0.6902	0.0298	0.0008	0.0039	0.2216	0.6764		6	0.0071	0.0000	0.6902	-0.0298	-0.0008	-0.0024	0.2216	0.6764	
		5	-0.0029	0.0001	-0.0004	-0.1997	0.7622	-0.0184	-0.0008	0.0024				5	-0.0029	0.0001	0.7622	-0.0184	-0.0008	0.0024			
57	2	7	0.798	-0.01	-0.00	0.0789	1.02	-1.00	-0.98	1.00	0.2321	0.7286	7	0.798	-0.01	0.0789	1.02	-1.00	-0.98	1.00	0.2321	0.7286	
		6	0.0083	0.0001	0.0011	0.0789	0.6650	0.0019	0.0000	0.0002	0.0539	0.6025		6	0.0083	0.0001	0.6650	0.0019	0.0000	0.0002	0.0539	0.6025	
		5	-0.0026	0.0001	-0.0003	-0.2626	0.7513	0.0073	0.0000	0.0008				5	-0.0026	0.0001	0.7513	0.0073	0.0000	0.0008			
57	3	7	0.797	0.46	-0.00	0.0801	1.02	-0.99	-0.98	1.01	0.0573	0.8366	7	0.797	0.46	0.0801	1.02	-0.99	-0.98	1.01	0.0573	0.8366	
		8	0.0079	0.0001	0.0010	0.0801	0.6618	0.0012	0.0002	0.0002	0.1130	0.6391		8	0.0079	0.0001	0.6618	0.0012	0.0002	0.0002	0.1130	0.6391	
		9	-0.0025	0.0001	-0.0004	-0.2240	0.8222	0.0119	0.0002	0.0002				9	-0.0025	0.0001	0.8222	0.0119	0.0002	0.0002			
57	4	7	0.799	1.00	-0.00	0.0889	1.02	-0.98	-0.98	1.01	0.3618	0.6967	7	0.799	1.00	0.0889	1.02	-0.98	-0.98	1.01	0.3618	0.6967	
		6	0.0084	0.0001	0.0011	0.0889	0.6585	0.0061	0.0004	0.0008	0.1186	0.6399		6	0.0084	0.0001	0.6585	0.0061	0.0004	0.0008	0.1186	0.6399	
		5	-0.0024	0.0001	-0.0003	-0.2925	0.7890	0.0162	0.0003	0.0020				5	-0.0024	0.0001	0.7890	0.0162	0.0003	0.0020			
57	5	7	0.799	2.06	-0.00	0.0947	1.02	-0.97	-0.98	1.02	0.2293	0.6558	7	0.799	2.06	0.0947	1.02	-0.97	-0.98	1.02	0.2293	0.6558	
		6	0.0080	0.0001	0.0010	0.0947	0.6801	0.0160	0.0007	0.0006	0.1282	0.6528		6	0.0080	0.0001	0.6801	0.0160	0.0007	0.0006	0.1282	0.6528	
		5	-0.0025	0.0001	-0.0004	-0.2016	0.8353	0.0256	0.0006	0.0034				5	-0.0025	0.0001	0.8353	0.0256	0.0006	0.0034			
57	6	7	0.798	5.09	-0.00	0.0982	1.02	-0.94	-0.98	1.04	0.1990	0.6586	7	0.798	5.09	0.0982	1.02	-0.94	-0.98	1.04	0.1990	0.6586	
		6	0.0075	0.0001	0.0010	0.0982	0.6729	0.0260	0.0010	0.0034	0.1270	0.6586		6	0.0075	0.0001	0.6729	0.0260	0.0010	0.0034	0.1270	0.6586	
		5	-0.0024	0.0001	-0.0003	-0.2920	0.7358	0.0363	0.0009	0.0047				5	-0.0024	0.0001	0.7358	0.0363	0.0009	0.0047			
57	7	7	0.798	5.13	-0.00	0.0816	1.02	-0.93	-0.98	1.05	0.1637	0.6526	7	0.798	5.13	0.0816	1.02	-0.93	-0.98	1.05	0.1637	0.6526	
		8	0.0070	0.0001	0.0009	0.0816	0.6646	0.0364	0.0013	0.0047	0.1333	0.6325		8	0.0070	0.0001	0.6646	0.0364	0.0013	0.0047	0.1333	0.6325	
		9	-0.0027	0.0001	-0.0003	-0.2499	0.6880	0.0457	0.0012	0.0057				9	-0.0027	0.0001	0.6880	0.0457	0.0012	0.0057			
57	8	7	0.798	-0.03	-0.00	0.0596	1.02	-1.00	-0.98	1.09	0.1644	0.8203	7	0.798	-0.03	0.0596	1.02	-1.00	-0.98	1.09	0.1644	0.8203	
		6	0.0071	0.0000	0.0009	0.0596	0.6686	0.0020	0.0000	0.0003	0.0560	0.5637		6	0.0071	0.0000	0.6686	0.0020	0.0000	0.0003	0.0560	0.5637	
		9	-0.0030	0.0001	-0.0004	-0.2344	0.7273	0.0080	0.0000	0.0009				9	-0.0030	0.0001	0.7273	0.0080	0.0000	0.0009			
58	1	7	0.799	-5.10	-0.00	0.1030	2.01	-2.04	-2.00	1.97	0.1363	0.6642	7	0.799	-5.10	0.1030	2.01	-2.04	-2.00	1.97	0.1363	0.6642	
		6	0.0128	0.0002	0.0016	0.1030	0.6549	-0.0354	-0.0009	0.0047	0.2427	0.6791		6	0.0128	0.0002	0.6549	-0.0354	-0.0009	0.0047	0.2427	0.6791	
		9	-0.0065	-0.0000	-0.0009	0.0320	0.7437	-0.0128	-0.0006	0.0017				9	-0.0065	-0.0000	0.7437	-0.0128	-0.0006	0.0017			

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																											
58	2	7	0.798	-0.0002	-0.0017	0.1099	2.01	-1.99	-2.00	2.01	-0.0002	2.01	0.0648	0.6349	0.0134	0.0002	0.0017	0.1099	2.01	-1.99	-2.00	2.01	-0.0002	2.01	0.0648	0.6349			
		8	0.0134	0.0002	0.0017	0.0288	0.6480	-0.0062	-0.0000	0.4800	0.0000	-0.0000	0.0648	0.6349															
		9	-0.0065	-0.0000	-0.0009	0.0288	0.7236	0.0123	0.0002	0.7236	0.0002	0.0015	0.1008	0.6349															
58	3	7	0.797	0.0003	0.0018	0.1176	2.01	-1.98	-2.00	2.01	0.0000	2.01	-0.1746	0.6364	0.0138	0.0003	0.0018	0.1176	2.01	-1.98	-2.00	2.01	-0.0003	2.01	0.7402	0.6364			
		8	0.0138	0.0003	0.0018	0.0364	0.6559	-0.0024	0.0000	0.6559	0.0000	-0.0003	0.1746	0.6364															
		9	-0.0063	-0.0000	-0.0009	0.0364	0.7315	0.0159	0.0003	0.7315	0.0003	0.0020	0.1130	0.6364															
58	4	7	0.798	1.01	-0.00	0.1157	2.01	-1.97	-2.00	2.01	0.0002	2.02	0.9935	0.6428	0.0135	0.0003	0.0017	0.1157	2.01	-1.97	-2.00	2.02	0.0001	2.02	0.9935	0.6428			
		8	0.0135	0.0003	0.0017	0.0278	0.6383	0.0012	0.0002	0.6383	0.0002	0.0026	0.1158	0.6428															
		9	-0.0065	-0.0000	-0.0009	0.0278	0.7273	0.0208	0.0004	0.7273	0.0004	0.0026	0.1158	0.6428															
58	5	7	0.798	2.09	-0.00	0.1210	2.01	-1.95	-2.01	2.03	0.0005	2.03	0.2747	0.6671	0.0129	0.0003	0.0017	0.1210	2.01	-1.95	-2.01	2.03	0.0013	2.03	0.2747	0.6671			
		8	0.0129	0.0003	0.0017	0.0145	0.6600	0.0103	0.0005	0.6600	0.0005	0.0042	0.1203	0.6671															
		9	-0.0063	-0.0000	-0.0009	0.0145	0.7142	0.0320	0.0007	0.7142	0.0007	0.0042	0.1203	0.6671															
58	6	7	0.798	3.11	-0.00	0.1133	2.00	-1.94	-2.01	2.04	0.0009	2.04	0.2168	0.6560	0.0123	0.0002	0.0015	0.1133	2.00	-1.94	-2.01	2.04	0.0027	2.04	0.2168	0.6560			
		8	0.0123	0.0002	0.0015	0.0185	0.6456	0.0208	0.0009	0.6456	0.0009	0.0055	0.1259	0.6560															
		9	-0.0061	-0.0000	-0.0009	0.0185	0.7339	0.0430	0.0010	0.7339	0.0010	0.0055	0.1259	0.6560															
58	7	7	0.799	4.11	-0.00	0.1143	2.00	-1.92	-2.01	2.06	0.0012	2.06	0.1909	0.6533	0.0126	0.0002	0.0015	0.1143	2.00	-1.92	-2.01	2.06	0.0041	2.06	0.1909	0.6533			
		8	0.0126	0.0002	0.0016	0.0113	0.6430	0.0319	0.0012	0.6430	0.0012	0.0063	0.1431	0.6533															
		9	-0.0063	-0.0000	-0.0009	0.0113	0.7578	0.0505	0.0014	0.7578	0.0014	0.0063	0.1431	0.6533															
58	8	7	0.798	-0.02	-0.00	0.1124	2.01	-1.99	-2.01	2.01	0.0000	2.01	0.0740	0.6262	0.0135	0.0003	0.0017	0.1124	2.01	-1.99	-2.01	2.01	0.0008	2.01	0.0740	0.6262			
		8	0.0135	0.0003	0.0017	0.0035	0.6459	-0.0057	0.0000	0.6459	0.0000	0.0015	0.1029	0.6262															
		9	-0.0065	-0.0000	-0.0009	0.0035	0.7217	0.0126	0.0002	0.7217	0.0002	0.0015	0.1029	0.6262															
59	1	7	0.798	-3.08	-0.00	0.1406	5.02	-5.07	-5.03	5.00	0.0015	5.00	0.1515	0.6415	0.0285	0.0008	0.0038	0.1406	5.02	-5.07	-5.03	5.00	0.0064	5.00	0.1515	0.6415			
		8	0.0285	0.0008	0.0038	0.1038	0.6828	-0.0503	0.0015	0.6828	0.0015	0.0064	0.2083	0.6415															
		9	-0.0219	-0.0004	-0.0030	0.1038	0.7023	0.0025	0.0001	0.7023	0.0001	0.0064	0.2083	0.6415															
59	2	7	0.799	-0.03	-0.00	0.1328	5.02	-5.02	-5.03	5.03	0.0005	5.03	0.1394	0.6687	0.0302	0.0008	0.0040	0.1328	5.02	-5.02	-5.03	5.03	0.0028	5.03	0.1394	0.6687			
		8	0.0302	0.0008	0.0040	0.1119	0.6715	-0.0210	0.0005	0.6715	0.0005	0.0036	0.1185	0.6687															
		9	-0.0219	-0.0004	-0.0030	0.1119	0.7008	0.0275	0.0006	0.7008	0.0006	0.0036	0.1185	0.6687															
59	3	7	0.799	0.48	-0.00	0.1336	5.02	-5.01	-5.03	5.04	0.0004	5.04	0.1460	0.7030	0.0303	0.0008	0.0041	0.1336	5.02	-5.01	-5.03	5.04	0.0022	5.04	0.1460	0.7030			
		8	0.0303	0.0008	0.0041	0.1096	0.6780	-0.0159	0.0004	0.6780	0.0004	0.0043	0.1180	0.7030															
		9	-0.0220	-0.0004	-0.0030	0.1096	0.6975	0.0326	0.0007	0.6975	0.0007	0.0043	0.1180	0.7030															
59	4	7	0.798	0.99	-0.00	0.1313	5.02	-5.00	-5.03	5.05	0.0002	5.05	0.1187	0.6920	0.0295	0.0007	0.0039	0.1313	5.02	-5.00	-5.03	5.05	0.0015	5.05	0.1187	0.6920			
		8	0.0295	0.0007	0.0039	0.1147	0.6767	-0.0115	0.0002	0.6767	0.0002	0.0050	0.1179	0.6920															
		9	-0.0219	-0.0005	-0.0031	0.1147	0.7095	0.0381	0.0008	0.7095	0.0008	0.0050	0.1179	0.6920															

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key														
59	5	7	0.798	2.07	-0.007	-0.00	0.1311	0.6688	5.02	-4.99	-5.03	5.06	-0.0377	0.7963			
		6	0.0293	0.0007	0.0039	0.1311	0.6688	5.02	-4.99	-5.03	5.06	-0.0377	0.7963				
		9	-0.0221	-0.0005	-0.0031	0.1152	0.7028	0.7028	-0.0035	0.0012	-0.0059	0.1311	0.6414				
59	6	7	0.798	3.14	-0.007	-0.00	0.1313	0.6767	5.01	-4.97	-5.03	5.08	0.3856				
		8	0.0294	0.0007	0.0039	0.1313	0.6767	5.01	0.0049	0.0003	0.0006	0.3856	0.6112				
		9	-0.0215	-0.0005	-0.0030	0.1173	0.7063	0.7063	0.0541	0.0015	0.0066	0.1435	0.6134				
59	7	7	0.798	4.12	-0.007	-0.00	0.1259	0.6299	5.01	-4.96	-5.03	5.08	0.2442				
		8	0.0295	0.0007	0.0039	0.1259	0.6299	5.01	0.0140	0.0006	0.0017	0.2442	0.6274				
		9	-0.0208	-0.0005	-0.0030	0.1250	0.7218	0.7218	0.0609	0.0016	0.0072	0.1355	0.5962				
59	8	7	0.799	-0.00	-0.00	-0.00	0.1320	0.6751	5.02	-5.02	-5.03	5.04	0.1441				
		8	0.0302	0.0007	0.0040	0.1320	0.6751	5.02	-0.0205	-0.0005	-0.0028	0.1441	0.6986				
		9	-0.0212	-0.0004	-0.0030	0.1080	0.7075	0.7075	0.0294	0.0007	0.0039	0.1240	0.6753				
60	1	7	0.799	-3.08	-0.00	-0.00	0.2970	-2.99	2.97	3.03	-3.07	0.0901	0.6477				
		8	0.0109	-0.0006	-0.0014	0.2970	-2.99	2.97	-0.0075	-0.0001	-0.0009	0.0901	0.6477				
		9	-0.0139	0.0008	0.0019	0.2901	0.6967	0.6967	-0.0399	-0.0015	-0.0052	0.1877	0.6628				
60	2	7	0.798	-0.05	-0.00	-0.00	0.3156	-2.99	3.02	3.03	-3.02	0.2198	0.6856				
		8	0.0095	-0.0006	-0.0013	0.3156	-2.99	3.02	0.0173	0.0007	0.0023	0.2198	0.7317				
		9	-0.0149	0.0008	0.0020	0.2749	0.6717	0.6717	-0.0118	-0.0006	-0.0017	0.2534	0.6956				
60	3	7	0.799	0.47	-0.00	-0.00	0.3025	-2.99	3.03	3.03	-3.02	0.1997	0.6768				
		8	0.0099	-0.0006	-0.0013	0.3025	-2.99	3.03	0.0235	0.0009	0.0031	0.1997	0.8220				
		9	-0.0149	0.0008	0.0020	0.2826	0.6853	0.6853	-0.0056	-0.0003	-0.0009	0.3496	0.6768				
60	4	7	0.798	0.99	-0.00	-0.00	0.2881	-2.99	3.04	3.03	-3.01	0.1889	0.6725				
		8	0.0096	-0.0005	-0.0013	0.2881	-2.99	3.04	0.0286	0.0010	0.0038	0.1889	1.0386				
		9	-0.0148	0.0008	0.0020	0.2812	0.6809	0.6809	-0.0027	-0.0003	-0.0005	0.6087	0.6725				
60	5	7	0.798	2.05	-0.00	-0.00	0.2785	-2.98	3.06	3.03	-3.00	0.1796	0.6570				
		8	0.0094	-0.0005	-0.0012	0.2785	-2.98	3.06	0.0388	0.0013	0.0051	0.1796	0.4531				
		9	-0.0151	0.0008	0.0020	0.2809	0.6799	0.6799	-0.0037	-0.0000	-0.0003	-0.0848	0.6570				
60	6	7	0.799	3.13	-0.00	-0.00	0.2797	-2.98	3.07	3.04	-2.99	0.1783	0.6368				
		8	0.0096	-0.0005	-0.0013	0.2797	-2.98	3.07	0.0476	0.0017	0.0060	0.1783	0.6085				
		9	-0.0150	0.0008	0.0020	0.2837	0.6853	0.6853	0.0124	0.0002	0.0015	0.1111	0.6085				
60	7	7	0.798	4.16	-0.00	-0.00	0.2736	-2.99	3.08	3.04	-2.97	0.1841	0.6132				
		8	0.0098	-0.0005	-0.0013	0.2736	-2.99	3.08	0.0539	0.0019	0.0066	0.1841	0.6319				
		9	-0.0156	0.0008	0.0021	0.2866	0.6914	0.6914	0.0225	0.0006	0.0028	0.1338	0.6319				

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																		
60	8	0.799	-0.0006	-0.00	0.2961	-2.99	3.02	3.04	-3.02	0.2141	0.6826	0.0101	-0.0008	-0.0013	0.2823	0.6734	0.0180	3.04	0.0024	0.7679
	8	-0.0149	0.0008	0.0020	0.0000	0.6809	-0.0106	-0.0005	-0.0016	0.2728	0.7679									
61	6	1.249	-3.12	0.00	0.0552	-2.98	2.99	3.02	-3.05	-0.0600	0.6710	-0.0142	-0.0001	-0.0019	0.0695	0.6789	-0.0082	0.0000	-0.0011	0.7001
	6	-0.0149	0.0002	0.0021	0.0000	0.7290	-0.0406	0.0001	-0.0056	-0.0179	0.7001									
61	7	1.248	-0.002	0.00	0.0723	-2.98	3.02	3.02	-3.03	0.0396	0.7007	-0.0132	-0.0001	-0.0023	0.0817	0.6739	0.0174	0.0001	0.0024	0.7445
	7	-0.0159	0.0002	0.0023	0.0000	0.7267	-0.0131	-0.0001	-0.0019	0.0715	0.7445									
61	8	1.253	0.48	0.00	0.0628	-2.98	3.03	3.02	-3.03	0.0277	0.6960	-0.0128	-0.0001	-0.0017	0.0675	0.6774	0.0222	0.0001	0.0030	0.7751
	8	-0.0164	0.0002	0.0023	0.0000	0.7265	-0.0094	-0.0002	-0.0014	0.1100	0.7751									
61	0	1.250	1.04	0.00	0.0615	-2.98	3.03	3.02	-3.02	0.0190	0.6952	-0.0126	-0.0001	-0.0017	0.0771	0.6936	0.0273	0.0001	0.0036	0.8620
	0	-0.0163	0.0002	0.0023	0.0000	0.7245	-0.0045	-0.0001	-0.0007	0.1961	0.8620									
61	10	1.243	2.08	0.00	0.0531	-2.98	3.04	3.02	-3.01	0.0012	0.6813	-0.0126	-0.0001	-0.0017	0.0897	0.6836	0.0369	0.0001	0.0050	0.3443
	10	-0.0161	0.0002	0.0023	0.0000	0.7390	0.0024	-0.0001	0.0001	-0.3465	0.3443									
61	11	1.244	3.14	0.00	0.0494	-2.98	3.05	3.02	-3.00	-0.0115	0.6835	-0.0128	-0.0001	-0.0023	0.0804	0.6844	0.0465	0.0001	0.0012	0.5913
	11	-0.0164	0.0002	0.0023	0.0000	0.7239	0.0108	-0.0001	0.0001	-0.0620	0.5913									
61	12	1.243	4.19	0.00	0.0532	-2.98	3.06	3.02	-3.05	-0.0200	0.6837	-0.0135	-0.0001	-0.0024	0.0924	0.6731	0.0542	0.0002	0.0052	0.2682
	12	-0.0164	0.0003	0.0024	0.0000	0.7460	0.0970	-0.0046	-0.0052	-0.2477	0.2682									
61	13	1.243	-0.004	0.00	0.0750	-2.99	3.02	3.02	-3.03	0.0315	0.6912	-0.0129	-0.0001	-0.0017	0.0953	0.6861	0.0175	0.0001	0.0024	0.7654
	13	-0.0158	0.0003	0.0022	0.0000	0.7251	-0.0135	-0.0001	-0.0020	0.0689	0.7654									
62	3	1.252	0.48	0.10	-0.8951	0.62	-0.48	-0.48	0.51	0.2575	0.6394	-0.0349	0.0062	0.0045	-3.2522	-0.6531	0.0037	0.0001	0.0007	0.5286
	3	-0.0005	0.0003	-0.0000	-0.0000	0.4848	0.0074	-0.0002	0.0007	-0.1370	0.5286									
62	4	1.250	1.03	0.00	-0.1539	0.53	-0.48	-0.48	0.51	0.1079	0.6570	-0.0044	0.0001	0.0005	-2.9049	-0.6517	0.0090	0.0001	0.0014	0.6085
	4	-0.0005	0.0003	-0.0000	-0.0000	0.8532	0.0115	-0.0001	0.0014	-0.0628	0.6085									

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key →															
62	5	7	1.250	-0.0001	2.08	-0.0003	-0.0001	0.0005	-0.1628	0.6839	0.52	-0.0179	-0.0002	0.0001	0.51	0.0025	-0.0505	0.6614
		8	0.0041	0.0003	-0.0001	-0.0001	0.0005	-2.5408	0.9542	0.6839	0.52	0.0179	0.0002	0.0001	0.0025	-0.0505	0.6355	
		9	-0.0006	0.0003	0.0003	-0.0001	-0.0001	-2.5408	0.9542	0.6839	0.52	0.0203	-0.0002	0.0001	0.0025	-0.0599	0.6355	
62	6	7	1.249	-0.0001	3.11	-0.0003	0.0004	-0.1524	0.6846	0.53	-0.046	-0.0003	0.0000	0.51	0.0036	0.0162	0.6686	
		8	0.0034	0.0003	-0.0001	-0.0001	0.0004	-2.7689	0.2599	0.6846	0.53	0.0273	0.0000	0.0000	0.0036	0.0162	0.6686	
		9	-0.0006	0.0003	0.0003	-0.0001	-0.0001	-2.7689	0.2599	0.6846	0.53	0.0293	-0.0003	0.0000	0.0036	-0.0555	0.6585	
62	7	7	1.247	-0.0001	4.20	-0.0003	0.0003	-0.2369	0.6296	0.52	-0.045	-0.0004	0.0000	0.52	0.0047	-0.0009	0.6708	
		8	0.0030	0.0001	-0.0001	-0.0001	0.0003	-5.4098	0.6760	0.6296	0.52	0.0354	-0.0004	0.0000	0.0047	-0.0009	0.6708	
		9	-0.0003	0.0003	0.0003	-0.0001	-0.0001	-5.4098	0.6760	0.6296	0.52	0.0386	-0.0004	0.0000	0.0051	-0.0097	0.6621	
62	8	7	1.245	-0.0001	0.04	-0.0003	0.0005	-0.1762	0.6531	0.53	-0.049	-0.0002	0.0001	0.49	0.0001	-1.2582	1.1944	
		8	0.0044	0.0001	-0.0001	-0.0001	0.0005	-2.1769	0.6234	0.6531	0.53	0.0005	0.0002	0.0001	0.0001	-0.3116	0.431	
		9	-0.0008	0.0003	0.0003	-0.0001	-0.0001	-2.1769	0.6234	0.6234	0.53	0.0038	-0.0002	0.0002	0.0003	-0.3116	0.431	
63	1	7	1.251	-0.0001	3.13	-0.0002	0.0008	-0.1045	0.6670	1.01	-1.02	-0.0295	-0.0002	0.98	0.0040	-0.0381	0.6900	
		8	0.0065	0.0001	-0.0001	-0.0001	0.0008	-0.4994	0.7338	1.01	0.295	0.0002	0.0001	0.0040	0.0027	-0.0440	0.7099	
		9	-0.0028	0.0002	0.0002	-0.0001	-0.0004	-0.4994	0.7338	1.01	0.195	-0.0001	0.0001	0.0027	0.0040	-0.0440	0.7099	
63	2	7	1.249	-0.0001	0.04	-0.0003	0.0009	-0.1117	0.6329	1.02	-0.098	-0.0032	-0.0002	1.00	0.0005	-0.2370	0.7837	
		8	0.0074	0.0001	-0.0001	-0.0001	0.0009	-0.5978	0.7309	1.02	0.329	0.0032	0.0001	0.0005	0.0006	-0.1992	0.5458	
		9	-0.0027	0.0003	0.0003	-0.0001	-0.0004	-0.5978	0.7309	1.02	0.060	-0.0002	-0.0002	0.0006	0.0006	-0.1992	0.5458	
63	3	7	1.248	-0.0001	0.47	-0.0003	0.0009	-0.1200	0.6484	1.02	-0.098	-0.0011	-0.0001	1.00	0.0001	0.8503	0.5876	
		8	0.0069	0.0001	-0.0001	-0.0001	0.0009	-0.6093	0.7460	1.02	0.011	0.0001	0.0001	0.0001	0.0012	-0.0957	0.6247	
		9	-0.0027	0.0003	0.0003	-0.0001	-0.0004	-0.6093	0.7460	1.02	0.103	-0.0001	-0.0001	0.0012	0.0012	-0.0957	0.6247	
63	4	7	1.248	-0.0001	0.00	-0.0003	0.0008	-0.2807	0.5898	1.02	-0.097	-0.0058	-0.0002	1.01	0.0007	0.1847	0.6509	
		8	0.0750	0.0002	-0.0002	-0.0001	0.0008	-0.5446	0.7373	1.02	0.058	0.0002	0.0001	0.0007	0.0007	-0.0679	3.0007	
		9	-0.0029	0.0003	0.0003	-0.0001	-0.0004	-0.5446	0.7373	1.02	0.141	-0.0001	-0.0001	0.0007	0.0007	-0.0679	3.0007	
63	5	7	1.247	-0.0001	1.05	-0.0003	0.0008	-0.2812	0.5915	1.02	-0.096	-0.0147	-0.0022	1.17	0.0019	0.0697	0.6577	
		8	0.0747	0.0002	-0.0002	-0.0001	0.0008	-0.5392	0.6700	1.02	0.047	0.0002	0.0001	0.0019	0.0019	-1.6045	-1.6530	
		9	-0.0028	0.0003	0.0003	-0.0001	-0.0003	-0.5392	0.6700	1.02	0.222	0.0001	0.0001	0.0019	0.0019	-1.6045	-1.6530	
63	6	7	1.248	-0.0001	3.14	-0.0003	0.0012	-0.9501	0.7238	1.11	-0.095	-0.0245	-0.0001	0.97	0.0032	0.0252	0.6647	
		8	0.0328	0.0002	-0.0002	-0.0001	0.0012	-0.5856	0.7005	1.11	0.1089	0.0001	0.0001	0.0032	0.0032	-0.2274	-0.0792	
		9	-0.0027	0.0003	0.0003	-0.0001	-0.0003	-0.5856	0.7005	1.11	0.049	-0.0001	-0.0001	0.0032	0.0032	-0.2274	-0.0792	
63	7	7	1.246	-0.0001	2.99	-0.0003	0.0009	-0.8901	0.5581	1.11	-0.095	-0.0325	-0.0000	0.97	0.0043	0.0053	0.6678	
		8	0.0365	0.0002	-0.0002	-0.0001	0.0009	-0.5983	0.7387	1.11	0.1185	0.0000	0.0000	0.0043	0.0043	-0.2136	-0.0885	
		9	-0.0027	0.0003	0.0003	-0.0001	-0.0004	-0.5983	0.7387	1.11	0.000	-0.0000	-0.0000	0.0043	0.0043	-0.2136	-0.0885	

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																
63	8	1.250	-0.0001	-0.0008	0.1235	0.6663	1.02	-0.0028	-0.0041	0.0004	-0.2076	0.0004	0.0491	0.0065	-0.0004	-0.0006	-0.1657	0.5420
	6	-0.0029	0.0003	-0.0004	-0.5996	0.6859	0.0063	0.0002	0.0002	0.0006	0.0002	0.0006	0.5420					
64	1	1.250	-0.0001	-0.0015	-0.0543	0.6706	-0.0347	-0.0002	-0.0002	1.98	-0.0405	0.0047	0.6897	0.0116	-0.0010	-0.0020	0.0909	0.7233
	6	-0.0071	0.0002	-0.0010	-0.1876	0.7365	-0.0138	0.0002	0.0002	0.0020	0.0002	0.0020	0.7233					
64	2	1.249	-0.0001	-0.0015	-0.0667	0.6611	-0.0076	-0.0001	-0.0001	2.01	-0.0701	0.0011	0.7247	0.0117	-0.0010	-0.0014	-0.0962	0.6380
	6	-0.0070	0.0003	-0.0010	-0.2241	0.7248	0.0110	0.0002	0.0002	0.0014	0.0002	0.0014	0.6380					
64	3	1.250	-0.0001	-0.0015	-0.0672	0.6560	-0.0033	-0.0001	-0.0001	2.01	-0.2101	0.0005	0.8058	0.0116	-0.0010	-0.0019	-0.0723	0.6529
	8	-0.0074	0.0003	-0.0010	-0.2178	0.7019	0.0148	0.0002	0.0002	0.0019	0.0002	0.0019	0.6529					
64	4	1.249	-0.0001	-0.0015	-0.0609	0.6606	-0.0013	-0.0001	-0.0001	2.01	0.7291	0.0001	0.4447	0.0116	-0.0010	-0.0025	-0.0643	0.6430
	6	-0.0071	0.0003	-0.0010	-0.2181	0.7357	0.0197	0.0002	0.0002	0.0025	0.0002	0.0025	0.4447					
64	5	1.249	-0.0001	-0.0015	-0.0612	0.6549	-0.0098	-0.0001	-0.0001	2.02	0.0986	0.0012	0.6330	0.0115	-0.0010	-0.0037	-0.0611	0.6641
	8	-0.0071	0.0002	-0.0010	-0.2008	0.7494	0.0294	0.0003	0.0003	0.0037	0.0003	0.0037	0.6330					
64	6	1.249	-0.0001	-0.0013	-0.0647	0.6426	-0.0097	-0.0001	-0.0001	2.03	0.0450	0.0025	0.6612	0.0105	-0.0010	-0.0050	-0.0605	0.6697
	9	-0.0073	0.0002	-0.0010	-0.1983	0.7370	0.0376	0.0004	0.0004	0.0050	0.0004	0.0050	0.6612					
64	7	1.248	-0.0001	-0.0013	-0.0707	0.6491	-0.0283	-0.0001	-0.0001	2.03	0.0136	0.0037	0.6664	0.0102	-0.0010	-0.0062	-0.0605	0.6654
	9	-0.0075	0.0002	-0.0010	-0.1941	0.7167	0.0468	0.0005	0.0005	0.0062	0.0005	0.0062	0.6654					
64	8	1.247	-0.0001	-0.0016	-0.0634	0.6657	-0.0076	-0.0001	-0.0001	2.01	-0.0533	0.0011	0.7677	0.0121	-0.0010	-0.0013	-0.1044	0.6312
	9	-0.0071	0.0003	-0.0010	-0.2428	0.7266	0.0106	0.0002	0.0002	0.0013	0.0002	0.0013	0.7677					
65	1	1.249	-0.0001	-0.0038	-0.0417	0.6994	-0.0510	-0.0004	-0.0004	4.99	-0.0438	0.0070	0.6891	0.0274	-0.0038	-0.0000	-1.1337	0.2260
	8	-0.0300	0.0016	-0.0087	-0.2697	1.4480	0.0013	0.0003	0.0003	0.0000	0.0003	0.0000	0.2260					
65	2	1.252	-0.0001	-0.0038	-0.0501	0.6957	-0.0270	-0.0001	-0.0001	5.00	-0.0386	0.0032	0.7160	0.0224	-0.0032	-0.0036	-0.0716	0.6743
	9	-0.0224	0.0003	-0.0032	-0.0859	0.7244	0.0270	0.0003	0.0003	0.0032	0.0003	0.0032	0.7160					

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key															
65	3	1.249	0.46	-0.002	0.0004	-0.0038	0.0509	0.6893	4.99	-0.184	-4.99	-5.01	5.01	-0.0042	-0.0026	-0.0384	0.7302
		0.0276	-0.0004	0.0002	0.0004	0.0032	-0.0885	0.7213	0.7213	0.0310	0.0004	0.0004	0.0004	0.0042	0.0026	-0.0668	0.6774
65	4	1.250	1.01	-0.002	0.0003	-0.0038	0.0495	4.99	4.99	-4.98	-5.01	5.01	5.01	0.0019	-0.0527	0.7361	
		0.0277	0.0003	0.0002	0.0003	0.0032	-0.0842	0.7212	0.7212	0.0353	0.0004	0.0004	0.0004	0.0047	-0.0675	0.6729	
65	5	1.252	2.08	-0.003	0.0003	-0.0037	0.0556	5.00	5.00	-4.97	-5.01	5.01	5.01	0.0007	-0.1585	0.8601	
		0.0271	0.0003	0.0003	0.0003	0.0033	-0.0817	0.7288	0.7288	0.0449	0.0005	0.0005	0.0005	0.0060	-0.0651	0.6731	
65	6	1.250	3.15	-0.002	0.0003	-0.0036	0.0514	4.99	4.99	-4.96	-5.02	5.02	5.02	0.0004	0.2506	0.5040	
		0.0266	0.0003	0.0002	0.0003	0.0033	-0.0778	0.7272	0.7272	0.0535	0.0006	0.0006	0.0006	0.0071	-0.0641	0.6649	
65	7	1.248	4.16	-0.002	0.0003	-0.0034	0.0574	4.99	4.99	-4.95	-5.01	5.01	5.01	0.0013	0.0934	0.6174	
		0.0252	0.0003	0.0002	0.0003	0.0033	-0.0791	0.7250	0.7250	0.0621	0.0007	0.0007	0.0007	0.0082	-0.0629	0.6616	
65	8	1.250	-0.002	0.0004	-0.0032	0.0038	0.0484	5.00	5.00	-4.99	-5.01	5.01	5.01	0.0035	-0.0335	0.7260	
		0.0280	0.0004	0.0002	0.0004	0.0032	-0.0918	0.7117	0.7117	0.0227	0.0003	0.0003	0.0003	0.0035	-0.0745	0.6721	
68	1	1.247	-3.10	-0.002	0.0002	-0.0016	0.0856	1.00	1.00	1.00	1.00	1.00	1.00	0.0014	-0.0679	0.7162	
		0.0127	0.0002	0.0002	0.0002	0.0013	-0.1017	0.6565	0.6565	-0.0107	0.0002	0.0002	0.0002	0.0015	0.1168	0.7211	
68	2	1.248	0.06	-0.001	0.0003	0.0008	0.1127	1.03	1.03	1.02	1.02	1.02	1.01	0.0008	0.1515	0.6591	
		0.0055	0.0003	0.0001	0.0003	0.0009	0.2674	0.7484	0.7484	0.0067	0.0001	0.0001	0.0001	0.0008	-0.1327	0.6309	
68	3	1.248	1.08	-0.001	0.0003	0.0015	0.0554	1.04	1.04	1.02	1.03	1.03	1.01	0.0016	0.0672	0.6722	
		0.0116	0.0003	0.0001	0.0003	0.0017	0.1343	0.6733	0.6733	0.0126	0.0002	0.0002	0.0002	0.0016	-0.0787	0.6508	
68	4	1.251	3.15	-0.001	0.0002	0.0034	0.0349	1.05	1.05	1.04	1.05	1.05	1.03	0.0034	0.0200	0.6820	
		0.0248	0.0002	0.0001	0.0002	0.0035	0.0392	0.7032	0.7032	0.0258	0.0003	0.0003	0.0003	0.0034	-0.0637	0.6708	
68	5	1.247	6.33	-0.004	0.0000	0.0059	0.0472	1.07	1.07	1.06	1.06	1.06	1.03	0.0061	-0.0220	0.6799	
		0.0434	0.0000	0.0004	0.0000	0.0064	-0.0055	0.6853	0.6853	0.0433	0.0001	0.0001	0.0001	0.0061	-0.0612	0.6702	

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key →															
68	6	7	1.251	7.49	44.99	-0.0536	1.09	0.6837	1.07	0.595	-0.0003	1.09	0.080	-0.0325	0.6745			
		8	0.0589	-0.0006	0.0080	-0.0187	0.6837	0.0651	0.0651	-0.0007	0.0085	0.0085	-0.0567	0.6593				
		9	0.0641	-0.0002	0.0087	-0.0187	0.6783	0.0651	-0.0007	0.0085	-0.0567	0.6593						
68	7	7	1.249	12.69	44.99	-0.0564	1.11	0.733	1.09	0.751	-0.0006	1.11	1.06	-0.0428	0.6640			
		8	0.0736	-0.0008	0.0099	-0.0222	0.6614	0.0840	0.0840	-0.0009	0.0109	0.0109	-0.0543	0.6500				
		9	0.0815	-0.0003	0.0107	-0.0222	0.6614	0.0840	-0.0009	0.0109	-0.0543	0.6500						
68	8	7	1.252	0.05	44.99	-0.1278	1.03	0.7281	1.02	0.073	0.0001	1.03	1.01	0.1265	0.6479			
		8	0.0056	-0.0001	0.0008	0.2614	0.7166	0.0068	0.0068	-0.0001	0.0008	0.0008	-0.1371	0.6000				
		9	0.0064	0.0003	0.0009	0.2614	0.7166	0.0068	-0.0001	0.0008	-0.1371	0.6000						
69	1	7	1.251	-5.07	44.99	0.4236	2.96	0.745	2.98	0.000	0.0001	2.98	2.98	-19.3929	2.0724			
		8	0.0026	-0.0002	-0.0002	-1.2017	0.3283	-0.0012	-0.0003	-0.0002	-0.0002	-1.2543	1.1516					
		9	-0.0009	0.0002	-0.0000	-1.2017	0.3283	-0.0012	-0.0003	-0.0002	-0.0002	-1.2543	1.1516					
69	2	7	1.248	0.05	44.99	-0.0490	2.99	0.7000	3.00	0.175	0.0001	3.01	3.01	0.0392	0.6746			
		8	0.0161	-0.0001	0.0022	0.0789	0.7181	0.0164	0.0164	-0.0002	0.0021	0.0021	-0.0691	0.6637				
		9	0.0163	0.0002	0.0023	0.0789	0.7181	0.0164	-0.0002	0.0021	0.0021	-0.0691	0.6637					
69	3	7	1.246	1.08	44.99	-0.0444	2.99	0.6972	3.01	0.227	0.0000	3.02	3.01	0.0135	0.6807			
		8	0.0225	-0.0002	0.0031	0.0404	0.7080	0.0232	0.0232	-0.0002	0.0031	0.0031	-0.0605	0.6688				
		9	0.0232	0.0001	0.0032	0.0404	0.7080	0.0232	-0.0002	0.0031	0.0031	-0.0605	0.6688					
69	4	7	1.250	3.20	44.99	-0.0469	3.01	0.6929	3.02	0.352	-0.0000	3.03	3.02	-0.0108	0.6829			
		8	0.0359	-0.0003	0.0049	0.0040	0.6982	0.0365	0.0365	-0.0004	0.0048	0.0048	-0.0680	0.6709				
		9	0.0372	0.0000	0.0051	0.0040	0.6982	0.0365	-0.0004	0.0048	0.0048	-0.0680	0.6709					
69	5	7	1.246	6.28	44.99	-0.0528	3.02	0.6859	3.04	0.537	-0.0003	3.05	3.03	-0.0339	0.6753			
		8	0.0533	-0.0005	0.0073	-0.0170	0.6843	0.0556	0.0556	-0.0006	0.0074	0.0074	-0.0607	0.6665				
		9	0.0571	-0.0001	0.0078	-0.0170	0.6843	0.0556	-0.0006	0.0074	0.0074	-0.0607	0.6665					
69	6	7	1.252	9.51	44.99	-0.0586	3.04	0.6766	3.05	0.695	-0.0005	3.08	3.04	-0.0419	0.6663			
		8	0.0687	-0.0008	0.0093	-0.0230	0.6650	0.0755	0.0755	-0.0008	0.0092	0.0092	-0.0562	0.6562				
		9	0.0751	-0.0003	0.0099	-0.0230	0.6650	0.0755	-0.0008	0.0092	0.0092	-0.0562	0.6562					
69	7	7	1.252	12.69	44.99	-0.0614	3.05	0.6638	3.06	0.835	-0.0007	3.10	3.06	-0.0478	0.6581			
		8	0.0825	-0.0010	0.0109	-0.0237	0.6481	0.0934	0.0934	-0.0010	0.0120	0.0120	-0.0552	0.6430				
		9	0.0911	-0.0004	0.0118	-0.0237	0.6481	0.0934	-0.0010	0.0120	0.0120	-0.0552	0.6430					
69	8	7	1.252	0.03	44.99	-0.0575	2.99	0.6856	3.00	0.178	0.0001	3.01	3.01	0.0320	0.6667			
		8	0.0159	-0.0001	0.0021	0.0852	0.7225	0.0166	0.0166	-0.0002	0.0021	0.0021	-0.0703	0.6555				
		9	0.0160	0.0002	0.0023	0.0852	0.7225	0.0166	-0.0002	0.0021	0.0021	-0.0703	0.6555					

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																	
70	1	1.251	-3.07	44.99	-0.0682	5.98	6.02	6.01	6.00	0.0021	0.0336	0.7006	0.0121	-0.0001	0.0001	0.0016	0.0021	0.0336	0.7006
	8	0.0121	-0.0002	0.0019	0.0865	0.7229	0.0126	0.0003	0.0000	0.0016	-0.1210	0.6646	0.0131	0.0002	0.0003	0.0016	0.0021	0.0336	0.7006
	9	0.0131	0.0002	0.0019	0.0865	0.7229	0.0126	0.0003	0.0000	0.0016	-0.1210	0.6646	0.0131	0.0002	0.0003	0.0016	0.0021	0.0336	0.7006
70	2	1.249	0.08	44.99	-0.0488	6.00	6.04	6.03	6.02	0.0046	-0.0090	0.6889	0.0318	-0.0003	0.0000	0.0046	0.0046	-0.0653	0.6889
	8	0.0318	-0.0003	0.0046	0.0017	0.7111	0.0329	-0.0004	0.0000	0.0046	-0.0653	0.6889	0.0327	0.0000	0.0004	0.0046	0.0046	-0.0653	0.6889
	9	0.0327	0.0000	0.0046	0.0017	0.7111	0.0329	-0.0004	0.0000	0.0046	-0.0653	0.6889	0.0327	0.0000	0.0004	0.0046	0.0046	-0.0653	0.6889
70	3	1.249	1.13	44.99	-0.0534	6.00	6.03	6.04	6.02	0.0055	-0.0238	0.6910	0.0385	-0.0004	0.0001	0.0055	0.0055	-0.0651	0.6910
	8	0.0385	-0.0004	0.0055	-0.0063	0.7024	0.0403	-0.0005	0.0000	0.0055	-0.0651	0.6910	0.0396	-0.0000	0.0005	0.0055	0.0055	-0.0651	0.6910
	9	0.0396	-0.0000	0.0055	-0.0063	0.7024	0.0403	-0.0005	0.0000	0.0055	-0.0651	0.6910	0.0396	-0.0000	0.0005	0.0055	0.0055	-0.0651	0.6910
70	7	1.253	-3.08	44.99	-0.0699	5.96	6.00	6.01	5.99	0.0017	0.0530	0.7440	0.0122	-0.0001	0.0001	0.0017	0.0022	0.0530	0.7440
	8	0.0122	-0.0001	0.0019	0.0758	0.7538	0.0122	0.0002	0.0000	0.0017	0.0530	0.7440	0.0130	0.0001	0.0002	0.0017	0.0022	0.0530	0.7440
	9	0.0130	0.0001	0.0019	0.0758	0.7538	0.0122	0.0002	0.0000	0.0017	0.0530	0.7440	0.0130	0.0001	0.0002	0.0017	0.0022	0.0530	0.7440
70	8	1.248	0.08	44.99	-0.0504	5.98	6.02	6.03	6.01	0.0046	-0.0022	0.6985	0.0319	-0.0003	0.0000	0.0046	0.0046	-0.0631	0.6985
	8	0.0319	-0.0003	0.0046	-0.0001	0.7229	0.0328	-0.0004	0.0000	0.0046	-0.0631	0.6985	0.0323	-0.0000	0.0004	0.0046	0.0046	-0.0631	0.6985
	9	0.0323	-0.0000	0.0046	-0.0001	0.7229	0.0328	-0.0004	0.0000	0.0046	-0.0631	0.6985	0.0323	-0.0000	0.0004	0.0046	0.0046	-0.0631	0.6985
70	9	1.247	1.14	44.99	-0.0521	5.99	6.02	6.04	6.02	0.0056	-0.0625	0.6997	0.0247	-0.0004	0.0005	0.0056	0.0054	-0.0625	0.6997
	8	0.0247	-0.0004	0.0056	-0.0087	0.7144	0.0400	-0.0005	0.0000	0.0056	-0.0625	0.6997	0.0285	-0.0004	0.0005	0.0056	0.0054	-0.0625	0.6997
	9	0.0285	-0.0004	0.0056	-0.0087	0.7144	0.0400	-0.0005	0.0000	0.0056	-0.0625	0.6997	0.0285	-0.0004	0.0005	0.0056	0.0054	-0.0625	0.6997
70	10	1.252	3.25	44.99	-0.0541	6.00	6.03	6.06	6.02	0.0074	-0.0298	0.6759	0.0514	-0.0005	0.0003	0.0074	0.0072	-0.0678	0.6759
	8	0.0514	-0.0005	0.0074	-0.0173	0.6951	0.0532	-0.0007	0.0000	0.0074	-0.0678	0.6759	0.0532	-0.0005	0.0007	0.0074	0.0072	-0.0678	0.6759
	9	0.0532	-0.0005	0.0074	-0.0173	0.6951	0.0532	-0.0007	0.0000	0.0074	-0.0678	0.6759	0.0532	-0.0005	0.0007	0.0074	0.0072	-0.0678	0.6759
70	11	1.253	6.37	44.99	-0.0596	6.01	6.05	6.08	6.04	0.0092	-0.0415	0.6754	0.0679	-0.0008	0.0005	0.0092	0.0095	-0.0618	0.6754
	8	0.0679	-0.0008	0.0092	-0.0220	0.6765	0.0722	-0.0008	0.0000	0.0092	-0.0618	0.6754	0.0725	-0.0003	0.0008	0.0092	0.0095	-0.0618	0.6754
	9	0.0725	-0.0003	0.0092	-0.0220	0.6765	0.0722	-0.0008	0.0000	0.0092	-0.0618	0.6754	0.0725	-0.0003	0.0008	0.0092	0.0095	-0.0618	0.6754
70	12	1.249	9.54	44.99	-0.0625	6.03	6.06	6.09	6.06	0.0109	-0.0564	0.6593	0.0826	-0.0010	0.0007	0.0109	0.0116	-0.0564	0.6593
	8	0.0826	-0.0010	0.0109	-0.0237	0.6598	0.0901	-0.0010	0.0000	0.0109	-0.0564	0.6593	0.0885	-0.0004	0.0010	0.0109	0.0116	-0.0564	0.6593
	9	0.0885	-0.0004	0.0109	-0.0237	0.6598	0.0901	-0.0010	0.0000	0.0109	-0.0564	0.6593	0.0885	-0.0004	0.0010	0.0109	0.0116	-0.0564	0.6593
70	13	1.250	12.73	44.99	-0.0611	6.04	6.07	6.11	6.06	0.0122	-0.0492	0.6465	0.0942	-0.0011	0.0009	0.0122	0.0134	-0.0556	0.6465
	8	0.0942	-0.0011	0.0122	-0.0288	0.6363	0.1072	-0.0011	0.0000	0.0122	-0.0492	0.6465	0.1044	-0.0006	0.0011	0.0122	0.0134	-0.0556	0.6465
	9	0.1044	-0.0006	0.0122	-0.0288	0.6363	0.1072	-0.0011	0.0000	0.0122	-0.0492	0.6465	0.1044	-0.0006	0.0011	0.0122	0.0134	-0.0556	0.6465
70	14	1.249	0.07	44.99	-0.0508	5.98	6.02	6.04	6.01	0.0046	-0.0074	0.6999	0.0321	-0.0003	0.0000	0.0046	0.0044	-0.0658	0.6999
	8	0.0321	-0.0003	0.0046	-0.0077	0.7270	0.0330	-0.0004	0.0000	0.0046	-0.0658	0.6999	0.0322	-0.0000	0.0000	0.0046	0.0044	-0.0658	0.6999
	9	0.0322	-0.0000	0.0046	-0.0077	0.7270	0.0330	-0.0004	0.0000	0.0046	-0.0658	0.6999	0.0322	-0.0000	0.0000	0.0046	0.0044	-0.0658	0.6999

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key											
71	1	7	1.250	-0.002	44.99	-0.0537	0.7185	0.0287	9.01	9.03	8.99	-0.0076	0.7044
		8	0.0277	0.0000	0.0039	0.0043	0.7330	0.0287	0.0313	-0.0000	0.0044	-0.0084	0.6794
		9	0.0282	0.0000	0.0041	0.0043	0.7330	0.0287	0.0287	-0.0004	0.0039	-0.0084	0.6794
71	2	7	1.250	0.10	44.99	-0.0551	0.6996	9.02	9.06	9.01	-0.0282	0.6879	
		8	0.0477	-0.0005	0.0066	-0.0199	0.7055	0.0488	0.0002	0.0067	0.0067	-0.0678	0.6776
		9	0.0481	-0.0001	0.0067	-0.0199	0.7055	0.0495	-0.0006	0.0067	0.0067	-0.0678	0.6776
71	3	7	1.249	1.13	44.99	-0.0576	0.6903	9.03	9.06	9.01	-0.0370	0.6870	
		8	0.0546	-0.0006	0.0075	-0.0203	0.7019	0.0544	-0.0004	0.0074	0.0074	-0.0640	0.6708
		9	0.0546	-0.0002	0.0076	-0.0203	0.7019	0.0561	-0.0007	0.0075	0.0075	-0.0640	0.6708
71	4	7	1.249	3.24	44.99	-0.0584	0.6776	9.04	9.08	9.02	-0.0428	0.6753	
		8	0.0665	-0.0007	0.0090	-0.0248	0.6811	0.0660	-0.0005	0.0089	0.0089	-0.0665	0.6622
		9	0.0681	-0.0003	0.0092	-0.0248	0.6811	0.0689	-0.0009	0.0091	0.0091	-0.0665	0.6622
71	5	7	1.250	6.41	44.99	-0.0617	0.6621	9.05	9.10	9.04	-0.0452	0.6608	
		8	0.0812	-0.0010	0.0108	-0.0247	0.6685	0.0819	-0.0007	0.0108	0.0108	-0.0602	0.6484
		9	0.0862	-0.0004	0.0114	-0.0247	0.6621	0.0862	-0.0010	0.0111	0.0111	-0.0602	0.6484
71	6	7	1.248	9.54	44.99	-0.0635	0.6424	9.09	9.12	9.05	-0.0484	0.6458	
		8	0.0943	-0.0011	0.0122	-0.0245	0.6510	0.0953	-0.0009	0.0123	0.0123	-0.0560	0.6287
		9	0.1009	-0.0004	0.0129	-0.0245	0.6424	0.1039	-0.0011	0.0130	0.0130	-0.0560	0.6287
71	7	7	1.250	12.72	44.99	-0.0615	0.6282	9.10	9.14	9.06	-0.0551	0.6387	
		8	0.1042	-0.0012	0.0133	-0.0270	0.6222	0.1062	-0.0011	0.0135	0.0135	-0.0551	0.6156
		9	0.1157	-0.0006	0.0145	-0.0270	0.6282	0.1191	-0.0012	0.0146	0.0146	-0.0551	0.6156
71	8	7	1.248	0.11	44.99	-0.0545	0.6966	9.04	9.06	9.01	-0.0304	0.6818	
		8	0.0480	-0.0005	0.0067	-0.0187	0.7055	0.0492	-0.0002	0.0067	0.0067	-0.0689	0.6713
		9	0.0482	-0.0001	0.0068	-0.0187	0.7055	0.0498	-0.0006	0.0066	0.0066	-0.0689	0.6713
72	1	7	1.249	-2.96	44.99	-0.0572	0.6901	15.05	15.05	15.02	-0.0352	0.6867	
		8	0.0619	-0.0007	0.0085	-0.0344	0.7019	0.0633	-0.0004	0.0084	0.0084	-0.0798	0.6794
		9	0.0585	-0.0004	0.0082	-0.0344	0.7019	0.0623	-0.0009	0.0084	0.0084	-0.0798	0.6794
72	2	7	1.250	0.16	44.99	-0.0547	0.6699	15.07	15.08	15.03	-0.0406	0.6683	
		8	0.0793	-0.0008	0.0106	-0.0324	0.6769	0.0791	-0.0006	0.0105	0.0105	-0.0628	0.6610
		9	0.0774	-0.0005	0.0104	-0.0324	0.6769	0.0809	-0.0010	0.0107	0.0107	-0.0628	0.6610
72	3	7	1.250	1.21	44.99	-0.0568	0.6643	15.07	15.08	15.04	-0.0431	0.6636	
		8	0.0846	-0.0009	0.0112	-0.0319	0.6700	0.0837	-0.0007	0.0111	0.0111	-0.0633	0.6523
		9	0.0824	-0.0005	0.0110	-0.0319	0.6700	0.0871	-0.0011	0.0113	0.0113	-0.0633	0.6523

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key														
72	4	1.249 0.0947 0.0939	5.29 -0.0010 -0.0005	44.99 0.0123 0.0123	15.08 0.6506 0.6558	15.07 0.0929 0.0981	15.09 -0.0007 -0.0012	15.04 0.0121 0.0125	-0.0429 -0.0627	0.6542 0.6395						
72	5	1.250 0.1063 0.1102	6.45 -0.0013 -0.0006	44.99 0.0135 0.0139	15.09 0.6364 0.6346	15.08 0.1060 0.1150	15.11 -0.0010 -0.0012	15.05 0.0135 0.0140	-0.0483 -0.0574	0.6398 0.6226						
72	6	1.250 0.1156 0.1215	9.59 -0.0015 -0.0006	44.99 0.0145 0.0150	15.10 0.6295 0.6187	15.09 0.1165 0.1255	15.13 -0.0013 -0.0013	15.06 0.0147 0.0152	-0.0584 -0.0547	0.6334 0.6064						
72	7	1.249 0.1236 0.1290	12.77 -0.0016 -0.0005	44.99 0.0154 0.0156	15.10 0.6262 0.6049	15.09 0.1234 0.1319	15.14 -0.0014 -0.0012	15.07 0.0155 0.0156	-0.0603 -0.0460	0.6282 0.5941						
72	8	1.249 0.0794 0.0773	0.16 -0.0008 -0.0004	44.99 0.0106 0.0104	15.06 0.6691 0.6751	15.06 0.0795 0.0813	15.08 -0.0006 -0.0010	15.04 0.0105 0.0106	-0.0409 -0.0656	0.6650 0.6560						
73	1	1.250 -0.0179 -0.0152	-3.14 0.0002 0.0002	44.99 -0.0024 -0.0020	0.01 0.6825 0.6771	-0.02 -0.0150 -0.0164	0.00 0.0000 -0.0002	-0.00 -0.0021 -0.0024	-0.0295 0.0638	0.7242 0.7470						
73	2	1.249 0.0009 0.0010	0.01 -0.0001 0.0003	44.99 0.0001 0.0002	0.04 0.7327 1.0628	-0.00 0.0014 0.0017	0.02 0.0001 -0.0002	0.00 0.0001 0.0000	0.5893 -0.6553	0.3802 0.0758						
73	3	1.249 0.0068 0.0063	1.05 -0.0001 0.0004	44.99 0.0008 0.0009	0.05 0.6536 0.7618	-0.00 0.0068 0.0073	0.03 0.0001 -0.0002	0.00 0.0008 0.0007	0.0990 -0.1484	0.6230 0.5265						
73	4	1.249 0.0200 0.0199	3.15 -0.0001 0.0003	44.99 0.0026 0.0027	0.06 0.6625 0.6985	0.00 0.0198 0.0202	0.04 0.0000 -0.0003	0.01 0.0025 0.0025	0.0241 -0.0744	0.6505 0.6199						
73	5	1.249 0.0382 0.0411	6.32 -0.0003 0.0000	44.99 0.0051 0.0056	0.08 0.775 0.6908	0.02 0.0381 0.0393	0.07 -0.0001 -0.0005	0.02 0.0050 0.0051	-0.0204 -0.0635	0.6682 0.6593						
73	6	1.248 0.0554 0.0590	9.48 -0.0005 -0.0001	44.99 0.0074 0.0080	0.10 0.753 0.6779	0.04 0.0553 0.0598	0.08 -0.0003 -0.0007	0.04 0.0073 0.0077	-0.0343 -0.0618	0.6655 0.6498						

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

See Key

Run	Pt.	Cd	1.250	12.65	44.99	-0.0556	0.6673	0.0102	0.05	0.704	0.0006	0.0093	-0.0429	0.6626
73	7	9	0.0702	-0.0007	0.0099	-0.0210	0.6611	0.0102	0.0704	-0.0006	0.0101	-0.0558	0.6444	
73	8	9	0.0013	-0.0001	0.0002	-0.6533	0.7335	0.0021	0.0024	0.0001	0.0000	0.3067	0.4025	
73	9	9	0.025	0.00	-0.00	-0.6564	0.04	-0.01	0.3926	-0.0267	0.0155	-0.5075	0.1970	
74	6	9	1.249	-3.15	44.99	-0.0010	0.6821	-3.02	-0.326	0.0003	-3.03	-0.0520	0.6893	
74	7	9	1.251	-0.001	44.99	0.0540	0.6554	-3.00	-0.0120	0.0001	-3.02	-0.0695	0.7206	
74	8	9	1.250	1.04	44.99	0.1141	0.6348	-2.99	-0.0081	0.0001	-3.02	-0.1111	0.261	
74	9	9	1.249	3.11	44.99	-0.1139	0.7228	-2.98	0.0025	-0.0002	-3.01	0.5071	0.5905	
74	10	9	1.250	6.28	44.99	-0.0212	0.6718	-2.96	0.0211	0.0001	-2.99	0.0470	0.6669	
74	11	9	1.250	9.43	44.99	-0.0354	0.6811	-2.94	0.0389	-0.0005	-2.95	-0.0003	0.6740	
74	12	9	1.250	12.61	44.99	-0.0415	0.6841	-2.92	0.0557	-0.0002	-2.93	-0.0248	0.6715	
74	13	9	1.250	-0.02	44.99	0.0598	0.6669	-3.00	-0.0132	0.0001	-3.01	-0.0636	0.7438	

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																									
75	1	7	1.048	-0.0141	0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	3.02	0.0002	-0.0011	-0.0067	-0.0011	-0.1203	0.6475	0.6829		
	8	9	-0.0184	0.1346	0.0814	0.6861	0.6952	-2.97	2.97	0.0090	-0.0495	0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001
75	2	7	1.048	-0.0132	0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	3.02	0.0002	-0.0028	-0.0022	0.0520	0.6863	0.7127		
	8	9	-0.0190	0.1267	0.0849	0.6735	0.6963	-2.97	2.97	0.0209	-0.0155	0.0002	-0.0002	-0.0002	-0.0002	-0.0002	-0.0002	-0.0002	-0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002
75	3	7	1.051	-0.0130	0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	3.02	0.0001	-0.0036	-0.0015	0.0344	0.6883	0.7262		
	8	9	-0.0191	0.1245	0.0823	0.6731	0.6935	-2.97	2.97	0.0266	-0.0103	0.0002	-0.0002	-0.0002	-0.0002	-0.0002	-0.0002	-0.0002	-0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002
75	4	7	1.048	-0.0129	0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	3.03	0.0001	-0.0043	-0.0009	0.0161	0.6806	0.7653		
	8	9	-0.0192	0.1242	0.0844	0.6763	0.6850	-2.97	2.97	0.0319	-0.0060	0.0002	-0.0002	-0.0002	-0.0002	-0.0002	-0.0002	-0.0002	-0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002
75	5	7	1.047	-0.0128	0.0002	-0.0002	-0.0002	-0.0002	-0.0002	-0.0002	-0.0002	-0.0002	-0.0002	-0.0002	-0.0002	-0.0002	-0.0002	-0.0002	-0.0002	3.03	0.0000	-0.0060	-0.0002	0.0000	0.6792	0.5145		
	8	9	-0.0190	0.1039	0.0942	0.6768	0.7055	-2.97	2.97	0.0446	-0.0026	0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
75	6	7	1.048	-0.0132	0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	3.03	0.0000	-0.0075	-0.0014	0.0072	0.6768	0.6158		
	8	9	-0.0192	0.0879	0.1039	0.6684	0.7112	-2.97	2.97	0.0557	-0.0121	0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
75	7	7	1.051	-0.0133	0.0002	-0.0002	-0.0002	-0.0002	-0.0002	-0.0002	-0.0002	-0.0002	-0.0002	-0.0002	-0.0002	-0.0002	-0.0002	-0.0002	-0.0002	3.03	0.0001	-0.0089	-0.0030	0.0133	0.6701	0.6373		
	8	9	-0.0199	0.0802	0.0940	0.6660	0.6924	-2.97	2.97	0.0669	-0.0239	0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	
75	8	7	1.046	-0.0131	0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	3.02	0.0002	-0.0028	-0.0022	0.0516	0.6850	0.7254		
	8	9	-0.0191	0.1300	0.0876	0.6717	0.6961	-2.97	2.97	0.0209	-0.0154	0.0002	-0.0002	-0.0002	-0.0002	-0.0002	-0.0002	-0.0002	-0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	
76	1	7	1.051	-0.0053	0.0002	-0.0002	-0.0002	-0.0002	-0.0002	-0.0002	-0.0002	-0.0002	-0.0002	-0.0002	-0.0002	-0.0002	-0.0002	-0.0002	-0.0002	0.47	0.0002	-0.0041	-0.0034	0.0432	0.6725	0.6795		
	8	9	-0.0009	-0.2058	-1.3845	0.6851	0.8930	0.55	-0.53	-0.507	-0.0252	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	
76	2	7	1.045	-0.0064	0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	0.47	0.0004	-0.0006	-0.0006	0.49	0.2426	0.6403		
	8	9	-0.0004	-0.1736	-3.2110	0.6499	1.0920	0.55	-0.50	-0.0004	-0.0052	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	
76	3	7	1.046	-0.0063	0.0002	-0.0002	-0.0002	-0.0002	-0.0002	-0.0002	-0.0002	-0.0002	-0.0002	-0.0002	-0.0002	-0.0002	-0.0002	-0.0002	-0.0002	0.47	0.0002	-0.0005	-0.0011	0.50	0.7046	0.6244		
	8	9	-0.0006	-0.1620	-2.3039	0.6413	1.3212	0.55	-0.49	-0.0042	-0.0088	-0.0002	-0.0002	-0.0002	-0.0002	-0.0002	-0.0002	-0.0002	-0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt. Cd ← See Key →

76	4	7	9	1.047	0.0065	-0.0004	0.0002	-0.0003	0.0002	-0.0001	-0.0008	0.54	0.6469	1.0920	-0.49	0.089	0.0137	-0.0001	-0.47	0.0002	0.0018	0.0012	-0.1549	0.0649	0.6830	0.6560
76	5	7	9	1.053	0.0062	-0.0001	0.0001	0.0003	2.05	-0.0008	-0.1530	0.55	0.6448	2.0355	-0.48	0.202	0.0240	-0.0002	-0.46	0.0002	0.0031	0.0027	0.0577	-0.0534	0.6678	0.6640
76	6	7	9	1.049	0.0060	-0.0001	0.0001	0.0003	3.09	-0.0007	-0.1604	0.55	0.6255	2.4279	-0.47	0.312	0.0351	-0.0003	-0.47	0.0001	0.0041	0.0046	0.0247	-0.0478	0.6724	0.6669
76	7	7	9	1.048	0.0056	-0.0002	0.0001	0.0003	4.15	-0.0007	-0.1573	0.55	0.6572	1.8360	-0.46	0.431	0.0474	-0.0004	-0.46	0.0000	0.0057	0.0063	0.0071	-0.0522	0.6712	0.6668
76	8	7	9	1.050	0.0064	-0.0006	0.0002	0.0003	-0.0001	-0.0008	-0.1783	0.55	0.6356	0.7970	-0.50	0.005	0.0051	-0.0001	-0.47	0.0002	0.0000	0.0005	0.49	-2.4838	0.5730	0.5649
77	3	7	9	1.048	0.0084	-0.0002	0.0001	0.0002	5.10	-0.0011	-0.1258	1.01	0.6725	0.8429	-1.04	0.351	0.0225	-0.0001	-0.99	0.0003	0.0047	0.0030	0.437	-0.0426	0.6705	0.6756
77	4	7	9	1.050	0.0084	-0.0024	0.0002	0.0002	0.06	-0.0011	-0.1191	1.01	0.6735	0.9530	-1.00	0.041	0.0073	-0.0001	-0.99	0.0002	1.01	0.0009	0.3127	-0.1107	0.6536	0.6511
77	5	7	9	1.048	0.0087	-0.0024	0.0001	0.0002	0.43	-0.0011	-0.0977	1.01	0.6812	0.8569	-1.00	0.005	0.0115	-0.0001	-0.99	0.0002	1.01	0.0000	2.8191	-0.0718	0.7711	0.6645
77	6	7	9	1.049	0.0087	-0.0026	0.0001	0.0002	1.01	-0.0011	-0.1114	1.01	0.6591	0.8282	-0.99	0.054	0.0174	-0.0002	-0.99	0.0003	1.01	0.0022	0.2958	-0.0574	0.7299	0.6366
77	7	7	9	1.047	0.0090	-0.0027	0.0001	0.0002	2.05	-0.0011	-0.0937	1.02	0.6446	0.7898	-0.98	0.172	0.0279	-0.0002	-0.99	0.0002	1.01	0.0023	0.0772	-0.0502	0.6731	0.6710
77	8	7	9	1.048	0.0083	-0.0022	0.0001	0.0002	3.13	-0.0010	-0.1159	1.01	0.6478	0.8680	-0.97	0.280	0.0403	-0.0003	-0.99	0.0002	1.02	0.0037	0.0387	-0.0481	0.6707	0.6664

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																				
77	9	1.047	4.16	-0.0001	-0.0003	-0.1115	1.01	-0.367	-0.99	1.03	0.0160	0.6700	0.0081	-0.0002	0.0010	0.6556	0.0387	0.0001	0.0051	0.0068	-0.0533	0.6664
	8	0.0022	0.0002	0.0002	0.0003	-0.6201	0.8153	0.0516	0.0005	0.0068	0.0160	0.6664	-0.0022	0.0002	-0.0003	0.8153	0.0516	0.0005	0.0068	0.0068	-0.0533	0.6664
77	10	1.049	-0.0002	0.0002	-0.0004	-0.1138	1.01	-1.01	-0.99	1.00	-0.3492	0.7598	0.0089	0.0002	0.0011	0.6662	-0.0032	0.0002	-0.0004	0.0009	-0.1062	0.6177
	8	0.0026	0.0002	0.0002	-0.0004	-0.4921	0.8375	0.0079	-0.0001	0.0009	-0.3492	0.7598	-0.0026	0.0002	-0.0004	0.8375	0.0079	-0.0001	0.0009	0.0009	-0.1062	0.6177
78	1	1.454	-3.13	-0.0001	-0.0010	-0.0635	2.00	-2.00	-2.00	1.97	-0.2066	-0.0723	0.0123	0.0001	0.0016	0.6974	-0.1015	0.0041	0.0014	0.0018	-0.0813	0.6800
	8	0.0074	0.0001	0.0001	-0.0010	-0.1283	0.6974	-0.0138	-0.0002	0.0014	-0.2066	-0.0723	0.0074	0.0001	-0.0010	0.6974	-0.0138	-0.0002	0.0014	0.0018	-0.0813	0.6800
78	2	1.046	-1.05	-0.0001	-0.0019	-0.0628	2.00	-5.37	-2.01	1.99	-0.1933	3.1768	0.0149	0.0001	0.0019	0.6461	-0.0128	0.0004	-0.0016	0.0016	-0.0697	0.6498
	8	0.0073	0.0001	0.0001	-0.0011	-0.1287	0.7762	0.0128	-0.0001	0.0016	-0.1933	3.1768	-0.0073	0.0001	-0.0011	0.7762	0.0128	-0.0001	0.0016	0.0016	-0.0697	0.6498
78	3	1.050	0.44	-0.0001	-0.0059	-0.2635	1.99	-1.98	-2.00	2.00	-0.2667	0.7291	0.0929	0.0001	0.0059	0.3183	-0.0042	0.0002	-0.0023	0.0023	-0.0487	0.6531
	8	0.0075	0.0001	0.0001	-0.0011	-0.1300	0.7892	0.0180	-0.0001	0.0023	-0.2667	0.7291	-0.0075	0.0001	-0.0011	0.7892	0.0180	-0.0001	0.0023	0.0023	-0.0487	0.6531
78	4	1.051	-0.01	0.0071	-0.0010	-1.2024	2.09	-1.98	-2.00	2.01	-0.7468	0.6563	0.0299	0.0002	0.0065	2.09	-0.0039	0.0005	-0.0028	0.0028	-0.0510	0.6563
	8	0.0073	0.0002	0.0002	-0.0010	-0.1489	0.7360	0.0220	-0.0002	0.0028	-0.7468	0.6563	-0.0073	0.0002	-0.0010	0.7360	0.0220	-0.0002	0.0028	0.0028	-0.0510	0.6563
78	5	1.047	1.04	-0.0001	-0.0019	-0.0552	2.00	-5.36	-2.01	2.01	-0.1312	0.6570	0.0148	0.0001	0.0019	0.6428	0.0108	0.0002	0.0014	0.0044	-0.0502	0.6570
	8	0.0076	0.0002	0.0002	-0.0011	-0.1486	0.7259	0.0336	-0.0003	0.0044	-0.1312	0.6570	-0.0076	0.0001	-0.0011	0.7259	0.0336	-0.0003	0.0044	0.0044	-0.0502	0.6570
78	6	1.048	3.12	-0.0001	-0.0018	-0.0599	2.00	-1.96	-2.00	2.01	-0.0601	0.6613	0.0146	0.0001	0.0018	0.6482	0.0218	0.0002	0.0028	0.0061	-0.0515	0.6613
	8	0.0081	0.0002	0.0002	-0.0011	-0.1524	0.6971	0.0452	-0.0004	0.0061	-0.0601	0.6613	-0.0081	0.0001	-0.0011	0.6971	0.0452	-0.0004	0.0061	0.0061	-0.0515	0.6613
78	7	1.049	4.16	-0.0001	-0.0018	-0.0671	2.00	-1.95	-2.00	2.02	-0.0236	0.6728	0.0143	0.0001	0.0018	0.6498	0.0325	0.0001	0.0043	0.0076	-0.0522	0.6728
	8	0.0076	0.0002	0.0002	-0.0010	-0.1706	0.6932	0.0571	-0.0005	0.0076	-0.0236	0.6728	-0.0076	0.0001	-0.0010	0.6932	0.0571	-0.0005	0.0076	0.0076	-0.0522	0.6728
78	8	1.048	-0.04	-0.0001	-0.0019	-0.0652	2.00	-1.99	-2.01	2.00	-0.1013	0.7019	0.0148	0.0001	0.0019	0.6458	-0.0090	0.0001	0.0012	0.0016	-0.0653	0.7019
	8	0.0077	0.0002	0.0002	-0.0011	-0.1419	0.7378	0.0128	-0.0001	0.0016	-0.1013	0.7019	-0.0077	0.0001	-0.0011	0.7378	0.0128	-0.0001	0.0016	0.0016	-0.0653	0.7019
79	1	1.049	-3.11	-0.0001	-0.0045	-0.0289	4.99	-5.05	-5.01	5.00	-0.0457	0.6785	0.0333	0.0001	0.0045	0.6858	-0.0618	0.0005	-0.0002	0.0002	-0.0457	0.6785
	8	0.0264	0.0003	0.0003	-0.0037	-0.0632	0.7027	0.0016	-0.0002	0.0002	-0.0457	0.6785	-0.0264	0.0001	-0.0037	0.7027	0.0016	-0.0002	0.0002	0.0002	-0.0457	0.6785

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key →																			
79	2	1.046	-0.002	-0.002	0.000	-0.0334	0.6861	4.99	-5.02	-0.0264	-5.01	5.02	-0.0373	0.6976							
	6	0.0337	-0.0002	-0.0002	0.0046	-0.0510	0.7107	0.0002	0.0327	0.0002	0.0001	-0.0044	-0.0448	0.6778							
	9	-0.0260	0.0002	0.0002	-0.0037	-0.0510	0.7107	0.0002	0.0327	0.0002	0.0001	-0.0044	-0.0448	0.6778							
79	3	1.046	0.002	0.002	-0.004	-0.0345	0.6843	4.99	-5.02	-0.0264	-5.01	5.02	-0.0373	0.6976							
	6	0.0338	-0.0002	-0.0002	0.0046	-0.0510	0.7107	0.0002	0.0327	0.0002	0.0001	-0.0044	-0.0448	0.6778							
	9	-0.0259	0.0002	0.0002	-0.0036	-0.0509	0.7096	0.0003	0.0385	-0.0003	0.0001	-0.0029	-0.0410	0.7052							
79	4	0.847	-0.003	-0.003	-0.000	-0.0357	0.6825	4.98	-5.01	-0.0255	-5.01	5.18	-0.0513	0.7093							
	6	0.0528	-0.0003	-0.0003	0.0072	-0.0526	0.7056	0.0002	0.0149	0.0125	0.0002	-0.0166	-4.1889	-5.5756							
	9	-0.0394	0.0004	0.0004	-0.0055	-0.0526	0.7056	0.0002	0.0149	0.0125	0.0002	-0.0166	-4.1889	-5.5756							
79	5	0.844	2.08	2.08	-0.10	-3.0351	5.07	-5.00	-5.00	-5.01	5.03	-0.0113	-0.1896	0.7910							
	6	0.0181	0.0110	0.0110	0.0141	-3.0478	3.8938	0.0866	0.0838	0.0003	0.0003	-0.0113	-0.0519	0.6779							
	9	-0.0399	0.0003	0.0003	-0.0057	-0.0478	0.7143	0.0866	0.0838	-0.0008	-0.0008	0.0113	-0.0519	0.6779							
79	6	1.048	3.11	3.11	-0.00	-0.0442	4.99	-4.99	-4.99	-5.02	5.04	0.0088	0.4135	0.5214							
	6	0.0333	-0.0002	-0.0002	0.0045	-0.0442	0.6835	0.0034	0.0654	0.0002	0.0003	0.0088	0.4135	0.5214							
	9	-0.0258	0.0002	0.0002	-0.0037	-0.0442	0.7171	0.0654	0.0654	-0.0007	0.0007	0.0088	0.4135	0.5214							
79	7	1.044	4.14	4.14	-0.00	-0.0470	4.97	-4.98	-4.98	-5.02	5.05	0.0166	-0.0971	0.6249							
	6	0.0322	-0.0003	-0.0003	0.0043	-0.0470	0.6806	0.0134	0.0766	0.0002	0.0002	0.0166	-0.0522	0.6689							
	9	-0.0262	0.0002	0.0002	-0.0037	-0.0445	0.7126	0.0766	0.0766	-0.0008	-0.0008	0.0166	-0.0522	0.6689							
79	8	1.046	-0.003	-0.003	-0.00	-0.0351	4.99	-5.02	-5.02	-5.01	5.03	-0.0043	-0.0348	0.7096							
	6	0.0338	-0.0002	-0.0002	0.0046	-0.0351	0.6806	0.0062	0.0324	0.0001	0.0001	-0.0043	-0.0445	0.6751							
	9	-0.0260	0.0002	0.0002	-0.0036	-0.0351	0.7043	0.0324	0.0324	-0.0002	-0.0002	-0.0043	-0.0445	0.6751							
80	1	0.897	-3.09	-3.09	-0.00	0.0805	5.00	-5.06	-5.06	-5.02	4.99	0.0002	-0.0953	0.6354							
	6	0.0318	0.0005	0.0005	0.0044	0.0514	0.6953	-0.0535	0.0020	-0.0010	-0.0068	0.0002	0.0953	0.6354							
	9	-0.0263	-0.0002	-0.0002	-0.0036	0.0514	0.6982	0.0020	0.0020	-0.0002	-0.0002	0.0002	0.0953	0.6354							
80	2	0.898	-0.005	-0.005	-0.00	0.0678	5.00	-5.03	-5.03	-5.02	5.02	-0.0045	-0.0800	0.7029							
	6	0.0331	0.0004	0.0004	0.0045	0.0628	0.6882	-0.0250	0.0333	-0.0004	-0.0035	0.0045	0.0800	0.7029							
	9	-0.0255	-0.0003	-0.0003	-0.0035	0.0628	0.6997	0.0333	0.0333	-0.0002	-0.0002	-0.0045	0.0413	0.6778							
80	3	0.898	0.45	0.45	-0.00	0.0663	5.00	-5.02	-5.02	-5.02	5.03	-0.0028	-0.0879	0.7192							
	6	0.0332	0.0004	0.0004	0.0046	0.0638	0.6917	-0.0196	0.0375	-0.0003	-0.0003	0.0028	0.0879	0.7192							
	9	-0.0255	-0.0003	-0.0003	-0.0036	0.0638	0.7059	0.0375	0.0375	-0.0003	-0.0003	-0.0028	0.0433	0.6797							
80	4	0.899	0.99	0.99	-0.00	0.0587	5.00	-5.01	-5.01	-5.02	5.04	-0.0058	-0.1114	0.7481							
	6	0.0335	0.0003	0.0003	0.0046	0.0675	0.6862	-0.0126	0.0435	-0.0002	-0.0018	0.0058	0.1114	0.7481							
	9	-0.0252	-0.0003	-0.0003	-0.0035	0.0675	0.7046	0.0435	0.0435	-0.0004	-0.0004	-0.0058	0.0552	0.6721							

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																				
80	5	7	0.899	2.05	-0.00	0.0691	5.00	-5.00	-5.02	5.05	0.0205	0.8396	0.0330	0.0004	0.0045	0.0725	0.6872	-0.0038	-0.0000	-0.0006	0.0723	0.6440	
		6	-0.0244	-0.0003	0.0045	0.0725	0.7233	0.0498	0.0007	0.0064													
		9			-0.0035	0.0725	0.7233	0.0498	0.0007	0.0064													
80	6	7	0.895	3.12	-0.00	0.0865	5.00	-4.99	-5.02	5.06	0.6888	0.5337	0.0311	0.0005	0.0042	0.0804	0.6771	0.0031	0.0004	0.0003	0.6804	0.6220	
		6	-0.0237	-0.0003	0.0042	0.0804	0.7033	0.0576	0.0009	0.0071													
		9			-0.0033	0.0804	0.7033	0.0576	0.0009	0.0071													
80	7	7	0.898	4.10	-0.00	0.0797	5.00	-4.98	-5.02	5.06	0.2088	0.6547	0.0306	0.0004	0.0042	0.0790	0.6909	0.0136	0.0005	0.0017	0.2088	0.6084	
		6	-0.0243	-0.0003	0.0042	0.0790	0.7117	0.0648	0.0009	0.0078													
		9			-0.0034	0.0790	0.7117	0.0648	0.0009	0.0078													
80	8	7	0.899	-0.04	-0.00	0.0632	5.00	-5.02	-5.02	5.03	0.0600	0.7157	0.0335	0.0004	0.0046	0.0496	0.6874	-0.0250	-0.0004	-0.0035	0.0600	0.6764	
		6	-0.0258	-0.0002	0.0046	0.0496	0.6881	0.0336	0.0002	0.0045													
		9			-0.0035	0.0496	0.6881	0.0336	0.0002	0.0045													
81	1	7	0.895	-3.08	-0.00	0.0767	2.00	-5.43	-5.34	1.99	0.0747	0.6808	0.0129	0.0001	0.0017	0.0675	0.6702	-0.0409	-0.0006	-0.0055	0.0747	0.6868	
		6	-0.0075	-0.0001	0.0009	0.0675	0.6526	-0.0156	-0.0005	0.0021													
		9			-0.0009	0.0675	0.6526	-0.0156	-0.0005	0.0021													
81	2	7	0.896	-0.26	-0.00	-0.2479	2.00	-5.38	-1.99	2.01	-0.2677	-0.4052	0.1070	-0.0053	-0.0073	-0.0515	0.3424	-0.0932	0.0049	0.0075	-0.2677	-0.4052	
		6	-0.0075	-0.0000	0.0009	-0.0515	0.6450	0.0125	0.0001	0.0107													
		9			-0.0009	-0.0515	0.6450	0.0125	0.0001	0.0107													
81	3	7	0.897	0.46	-0.00	0.0883	2.01	-2.00	-1.99	2.01	0.0747	0.6868	0.0139	0.0002	0.0018	0.0672	0.6576	-0.0183	-0.0001	-0.0024	0.0747	0.6868	
		6	-0.0073	-0.0000	0.0009	0.0672	0.6576	0.0183	0.0001	0.0024													
		9			-0.0009	0.0672	0.6576	0.0183	0.0001	0.0024													
81	4	7	0.897	1.03	-0.00	0.0918	2.01	-1.99	-1.99	2.02	1.9497	0.5409	0.0137	0.0002	0.0018	0.0824	0.6226	0.0008	0.0003	0.0032	1.9497	0.5409	
		6	-0.0066	-0.0001	0.0009	0.0824	0.7342	0.0245	0.0002	0.0032													
		9			-0.0009	0.0824	0.7342	0.0245	0.0002	0.0032													
81	5	7	0.898	2.00	88.79	0.1085	2.00	-1.98	-1.99	2.03	0.2834	0.6650	0.0128	0.0002	0.0017	0.0629	0.6833	0.0099	0.0005	0.0047	0.2834	0.6650	
		6	-0.0066	-0.0000	0.0009	0.0629	0.7298	0.0355	0.0002	0.0047													
		9			-0.0009	0.0629	0.7298	0.0355	0.0002	0.0047													
81	6	7	0.895	3.07	-0.00	0.1138	2.00	-1.96	-1.99	2.04	0.1678	0.6638	0.0124	0.0002	0.0016	0.0413	0.6747	0.0225	0.0007	0.0060	0.1678	0.6638	
		6	-0.0069	-0.0000	0.0009	0.0413	0.6915	0.0458	0.0005	0.0060													
		9			-0.0009	0.0413	0.6915	0.0458	0.0005	0.0060													
81	7	7	0.894	4.15	-0.00	0.1299	2.00	-1.95	-1.99	2.05	0.1256	0.6670	0.0117	0.0003	0.0016	0.0324	0.7049	0.0347	0.0008	0.0066	0.1256	0.6670	
		6	-0.0071	-0.0000	0.0009	0.0324	0.6867	0.0533	0.0008	0.0066													
		9			-0.0009	0.0324	0.6867	0.0533	0.0008	0.0066													

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key													
81	7	0.898	-0.002	-0.00	0.0911	0.6721	-2.00	-1.99	2.02	0.0720	0.7576				
	8	0.0140	0.0000	0.0018	0.0308	0.6561	-0.0142	-0.0000	-0.0009	0.0472	0.6483				
	9	-0.0073	-0.0000	-0.0009	0.0308	0.6561	0.0142	0.0000	0.0009						
82	7	0.896	-3.09	-0.00	0.0848	1.957	-1.02	-0.99	0.97	0.0684	0.6805				
	8	0.0072	0.0001	0.0010	0.1661	0.6123	-0.0345	-0.0004	-0.0047	0.1427	0.6891				
	9	-0.0037	0.0001	-0.0004	-0.1661	0.6123	0.0345	0.0006	-0.0031						
82	7	0.897	-0.05	-0.00	0.1032	1.03	-0.98	-0.99	1.00	-0.2617	0.6334				
	8	0.0078	0.0001	0.0010	0.2436	0.6847	-0.0028	0.0001	-0.0003	0.0407	0.6179				
	9	-0.0030	0.0001	-0.0003	-0.2436	0.6847	0.0028	0.0000	0.0003						
82	7	0.896	0.46	-0.00	0.1208	1.03	-0.98	-0.99	1.01	1.7242	0.6596				
	8	0.0080	0.0001	0.0011	0.3024	0.7125	0.0009	0.0003	0.0001	0.0570	0.6492				
	9	-0.0030	0.0001	-0.0003	-0.3024	0.7125	0.0122	0.0001	0.0015						
82	7	0.898	1.01	-0.00	0.1200	1.03	-0.97	-0.99	1.02	0.4819	0.6547				
	8	0.0076	0.0001	0.0010	0.2542	0.7026	0.0052	0.0005	0.0006	0.0415	0.6502				
	9	-0.0029	0.0001	-0.0003	-0.2542	0.7026	0.0185	0.0001	0.0024						
82	7	0.899	1.00	-0.00	0.1276	1.03	-0.96	-0.99	1.02	0.1923	0.6754				
	8	0.0073	0.0001	0.0009	0.2845	0.6623	0.0173	0.0006	0.0023	0.0358	0.6704				
	9	-0.0029	0.0001	-0.0003	-0.2845	0.6623	0.0296	0.0002	0.0039						
82	7	1.264	3.09	-0.00	0.1689	1.03	-0.95	-0.99	1.03	0.1366	0.6680				
	8	0.0049	0.0001	0.0007	0.3125	0.7201	0.0222	0.0006	0.0029	0.0425	0.6648				
	9	-0.0024	0.0001	-0.0002	-0.3125	0.7201	0.0326	0.0002	0.0043						
82	7	0.900	4.12	-0.00	0.1790	1.02	-0.94	-0.99	1.04	0.1159	0.6638				
	8	0.0056	0.0002	0.0008	0.3067	0.7036	0.0398	0.0009	0.0052	0.0667	0.6502				
	9	-0.0033	0.0002	-0.0003	-0.3067	0.7036	0.0495	0.0006	0.0064						
82	7	0.899	-0.03	-0.00	0.1029	1.03	-0.99	-0.99	1.00	-0.1902	0.6131				
	8	0.0078	0.0001	0.0010	0.2727	0.6589	-0.0078	0.0001	-0.0004	0.0080	0.5946				
	9	-0.0030	0.0001	-0.0003	-0.2727	0.6589	0.0078	0.0000	0.0009						
83	7	0.897	-3.08	-0.00	0.0109	0.53	-0.53	-0.48	0.48	0.0657	0.6786				
	8	0.0050	0.0000	0.0006	0.6183	0.6775	-0.0310	-0.0004	-0.0042	0.1339	0.6923				
	9	-0.0020	0.0002	-0.0002	-0.6183	0.6775	0.0254	-0.0006	-0.0035						
83	7	0.899	-0.05	-0.00	0.0960	0.53	-0.49	-0.48	0.51	-3.0748	0.5866				
	8	0.0051	0.0000	0.0007	0.9061	0.6919	-0.0004	0.0002	-0.0000	0.0076	0.5599				
	9	-0.0014	0.0002	-0.0001	-0.9061	0.6919	0.0058	-0.0001	0.0006						

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key															
83	5	7	0.897	0.000	0.45	-0.00	0.0007	0.0952	0.53	-0.48	-0.48	0.0004	0.51	0.7253	0.6787		
		6	0.0051	0.0000	0.0002	0.0000	0.0000	0.0952	0.6736	0.0032	0.0004	0.0004	0.0004	0.0455	0.6045		
		5	-0.0014	0.0000	0.0002	-0.0001	-0.0001	-0.9081	0.5166	0.0099	0.0000	0.0012	0.0012				
83	4	7	0.898	0.001	1.00	-0.00	0.0006	0.1043	0.53	-0.47	-0.48	0.51	0.3604	0.6915			
		6	0.0049	0.0001	0.0002	0.0000	0.0006	0.1043	0.6680	0.0080	0.0005	0.0011	0.0557	0.6546			
		5	-0.0015	0.0002	0.0002	-0.0001	-0.0001	-0.9054	0.4991	0.0140	0.0001	0.0018					
83	5	7	0.897	0.001	2.03	-0.00	0.0006	0.1170	0.53	-0.46	-0.49	0.52	0.1815	0.6889			
		6	0.0048	0.0001	0.0002	0.0006	0.0006	0.1170	0.6626	0.0210	0.0007	0.0029	0.0451	0.6664			
		5	-0.0018	0.0002	0.0002	-0.0001	-0.0001	-0.6942	0.3817	0.0266	0.0002	0.0035					
83	6	7	0.899	0.001	3.08	-0.00	0.0006	0.1985	0.53	-0.45	-0.48	0.53	0.1249	0.6763			
		6	0.0041	0.0002	0.0002	0.0006	0.0006	0.1985	0.7442	0.0327	0.0008	0.0044	0.0288	0.6661			
		5	-0.0013	0.0002	0.0002	-0.0001	-0.0001	-1.0741	0.4736	0.0379	0.0002	0.0050					
83	7	7	0.898	0.001	4.11	-0.00	0.0005	0.1771	0.53	-0.44	-0.48	0.54	0.1196	0.6591			
		6	0.0039	0.0001	0.0002	0.0005	0.0005	0.1771	0.6584	0.0418	0.0010	0.0055	0.0562	0.6592			
		5	-0.0009	0.0002	0.0002	-0.0000	-0.0000	-1.5844	0.4622	0.0479	0.0005	0.0063					
83	8	7	0.899	0.001	0.02	-0.00	0.0007	0.1039	0.53	-0.49	-0.48	0.50	-8.1974	1.2152			
		6	0.0052	0.0001	0.0003	0.0007	0.0007	0.1039	0.6844	-0.0001	0.0002	-0.0000	-0.0467	0.5956			
		5	-0.0012	0.0003	0.0003	-0.0001	-0.0001	-1.2306	0.5815	0.0060	-0.0000	-0.0007					
83	9	7	0.898	0.000	3.08	-0.00	0.0017	0.2875	-3.01	2.99	3.02	-3.06	0.0554	0.7135			
		6	0.0124	0.0007	0.0007	0.0017	0.0017	0.2875	0.6861	-0.0081	-0.0000	-0.0011	0.1294	0.6531			
		5	-0.0151	0.0007	0.0007	-0.0021	0.0021	0.2609	0.7208	-0.0437	-0.0011	-0.0057					
83	10	7	0.898	0.000	0.02	-0.00	0.0015	0.2874	-3.00	3.02	3.03	-3.03	0.1818	0.6906			
		6	0.0112	0.0006	0.0007	0.0015	0.0015	0.2874	0.6882	0.0198	0.0007	0.0027	0.2304	0.7458			
		5	-0.0174	0.0007	0.0007	-0.0024	0.0024	0.2236	0.6942	-0.0131	-0.0006	-0.0019					
84	1	7	0.900	0.000	3.07	-0.00	0.0017	0.2982	-2.97	2.98	3.03	-3.06	0.0346	0.6985			
		6	0.0120	0.0007	0.0007	0.0017	0.0017	0.2982	0.7079	-0.0080	-0.0000	-0.0011	0.1243	0.6558			
		5	-0.0159	0.0007	0.0007	-0.0022	0.0022	0.2419	0.7173	-0.0439	-0.0010	-0.0057					
84	2	7	0.900	0.000	0.02	-0.00	0.0015	0.2837	-2.96	3.03	3.03	-3.02	0.1783	0.6925			
		6	0.0113	0.0006	0.0007	0.0015	0.0015	0.2837	0.6955	0.0203	0.0007	0.0028	0.2246	0.7508			
		5	-0.0170	0.0007	0.0007	-0.0024	0.0024	0.2261	0.7074	-0.0128	-0.0005	-0.0019					
84	3	7	0.900	0.000	0.46	-0.00	0.0015	0.2866	-2.96	3.03	3.03	-3.02	0.1482	0.6938			
		6	0.0109	0.0006	0.0007	0.0015	0.0015	0.2284	0.7003	0.0256	0.0007	0.0035	0.3055	0.8048			
		5	-0.0171	0.0007	0.0007	-0.0024	0.0024	0.2284	0.7055	-0.0084	-0.0005	-0.0013					

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																											
84	4	7	8	9	0.400	-0.0109	-0.0168	1.00	-0.0007	0.00	-0.0015	0.00	0.2807	0.2336	-2.94	0.6986	0.7026	-0.0031	3.04	0.323	0.0008	3.03	-3.01	0.0044	0.0006	0.1265	0.5696	0.6864	1.0122
84	5	7	8	9	0.901	-0.0102	-0.0168	2.05	-0.0008	0.00	-0.0014	0.00	0.2721	0.2392	-2.96	0.7004	0.7081	0.0060	3.05	0.422	0.0010	3.03	-3.00	0.0055	0.0006	-0.1289	-0.0837	0.5582	0.5566
84	6	7	8	9	0.900	-0.0117	-0.0163	3.08	-0.0008	0.00	-0.0016	0.00	0.2576	0.2563	-2.96	0.6962	0.7195	0.0152	3.06	0.495	0.0013	3.03	-3.00	0.0063	0.0019	0.1369	0.1398	0.6368	0.6390
84	7	7	8	9	0.899	-0.0113	-0.0133	4.15	-0.0008	0.00	-0.0015	0.00	0.2547	0.2616	-2.96	0.7037	0.7168	0.0277	3.07	0.589	0.0014	3.03	-2.99	0.0073	0.0036	0.1222	0.1220	0.6202	0.6591
84	8	7	8	9	0.900	-0.0108	-0.0171	-0.002	-0.0007	-0.00	-0.0015	-0.00	0.2911	0.2319	-2.96	0.6930	0.7096	-0.0146	3.03	0.194	0.0007	3.03	-3.03	0.0026	-0.0021	0.1819	0.2066	0.6864	0.7392
85	3	7	8	9	0.898	-0.0396	-0.0394	-3.11	-0.0001	44.99	-0.0051	-0.0054	0.1420	0.0205	-3.00	0.6525	0.6847	-0.0392	-3.03	0.393	-0.0005	-3.04	-3.05	-0.0053	-0.0052	0.0739	0.0867	0.6847	0.6660
85	4	7	8	9	1.265	-0.0122	-0.0115	-0.00	-0.0001	44.99	-0.0015	-0.0016	0.2186	0.0465	-2.97	0.6249	0.7002	-0.0124	-3.01	0.137	-0.0001	-3.02	-3.02	-0.0019	-0.0017	0.0576	0.1561	0.7051	0.7159
85	5	7	8	9	0.898	-0.0083	-0.0066	1.02	-0.0000	44.99	-0.0010	-0.0010	0.2818	0.0718	-2.96	0.6360	0.7692	-0.0064	-3.00	0.082	-0.0000	-3.01	-3.01	-0.0011	-0.0010	0.0580	0.3204	0.7155	0.7984
85	6	7	8	9	0.899	-0.0029	-0.0042	3.06	0.0004	44.99	0.0005	0.0006	0.2178	0.5650	-2.94	0.9371	0.7566	0.0047	-2.98	0.004	0.0004	-2.99	-3.00	0.0000	0.0005	4.9538	-0.0940	0.7721	0.6070
85	7	7	8	9	0.900	-0.0273	-0.0295	6.19	0.0007	44.99	0.0036	0.0040	0.0782	0.1301	-2.91	0.6706	0.6853	0.0272	-2.95	0.230	0.0008	-2.96	-2.98	0.0030	0.0036	0.1871	0.0393	0.6651	0.6690
85	8	7	8	9	0.900	-0.0436	-0.0485	9.30	0.0012	44.99	0.0058	0.0061	0.1149	0.1318	-2.89	0.6672	0.6374	0.0524	-2.93	0.0419	0.0012	-2.93	-2.96	0.0054	0.0070	0.1500	0.0579	0.6488	0.6677

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key															
85	9	7	0.899	12.44	44.99	0.0013	0.0013	0.0074	0.1126	0.6093	-2.87	-2.91	-2.92	-2.94	0.1321	0.6302		
		8	0.0581	0.0013	0.0074	0.0013	0.1030	0.6093	0.0691	0.0580	0.0015	0.0008	0.0073	0.0638	0.6041			
		9	0.0653	0.0013	0.0079	0.0013	0.1030	0.6093	0.0691	0.0580	0.0008	0.0063	0.0063	0.0638	0.6041			
85	10	7	0.900	-0.01	44.99	-0.0006	0.0238	0.6970	-3.01	-3.02	-3.02	-3.02	0.0702	0.7246				
		8	-0.0157	-0.0000	44.99	-0.0020	0.0238	0.6970	-0.0151	-0.0004	-0.0022	-0.0022	0.1650	0.7384				
		9	-0.0149	-0.0000	44.99	-0.0020	0.0238	0.6970	-0.0151	-0.0004	-0.0022	-0.0022	0.1650	0.7384				
86	1	7	0.899	-3.10	44.99	-0.0001	0.1946	0.6563	-0.02	-0.02	0.00	-0.02	0.0683	0.6927				
		8	-0.0188	-0.0007	44.99	-0.0024	0.0487	0.6563	-0.0179	-0.0006	-0.0024	-0.0024	0.1762	0.6845				
		9	-0.0173	-0.0001	44.99	-0.0022	0.0487	0.6563	-0.0179	-0.0006	-0.0024	-0.0024	0.1762	0.6845				
86	2	7	0.898	0.00	44.99	0.0003	-0.1992	0.9690	-0.00	0.03	0.03	-0.00	-7.2083	1.4836				
		8	0.0019	-0.0000	44.99	0.0003	-1.0153	0.9441	-0.0032	-0.0001	-0.0004	-0.0004	-0.2618	0.6601				
		9	0.0018	0.0003	44.99	0.0003	-1.0153	0.9441	-0.0032	-0.0001	-0.0004	-0.0004	-0.2618	0.6601				
86	3	7	0.899	1.02	44.99	0.0010	0.1577	0.7977	0.00	0.04	0.04	0.01	0.5108	0.6837				
		8	0.0062	0.0001	44.99	0.0012	0.3734	0.7480	0.0057	0.0005	0.0001	0.0007	0.0849	0.6662				
		9	0.0082	0.0006	44.99	0.0012	0.3734	0.7480	0.0088	0.0001	0.0001	0.0011	0.0849	0.6662				
86	4	7	0.898	3.11	44.99	0.0008	0.0678	0.6869	0.01	0.06	0.06	0.02	0.1871	0.6711				
		8	0.0241	0.0003	44.99	0.0033	0.1757	0.7006	0.0195	0.0002	0.0002	0.0026	0.0555	0.6711				
		9	0.0240	0.0008	44.99	0.0033	0.1757	0.7006	0.0233	0.0002	0.0002	0.0031	0.0555	0.6711				
86	5	7	0.898	6.21	44.99	0.0013	0.0830	0.6713	0.04	0.09	0.09	0.03	0.1202	0.6509				
		8	0.0462	0.0007	44.99	0.0060	0.1472	0.6533	0.0419	0.0010	0.0006	0.0054	0.0695	0.6748				
		9	0.0460	0.0013	44.99	0.0060	0.1472	0.6533	0.0460	0.0006	0.0006	0.0062	0.0695	0.6748				
86	6	7	0.899	9.34	44.99	0.0014	8.1663	0.7680	0.06	0.11	0.11	0.06	0.1139	0.6263				
		8	0.0060	0.0099	44.99	0.0077	0.1172	0.6175	0.0585	0.0013	0.0010	0.0073	0.0806	0.6114				
		9	0.0627	0.0014	44.99	0.0077	0.1172	0.6175	0.0634	0.0010	0.0010	0.0077	0.0806	0.6114				
86	7	7	0.897	12.45	44.99	0.0014	0.0834	0.6146	0.08	0.12	0.12	0.06	0.0951	0.6070				
		8	0.0734	0.0012	44.99	0.0091	0.0927	0.5956	0.0734	0.0013	0.0010	0.0089	0.0662	0.5850				
		9	0.0766	0.0014	44.99	0.0091	0.0927	0.5956	0.0776	0.0010	0.0010	0.0090	0.0662	0.5850				
86	8	7	0.899	-0.01	44.99	-0.0003	-0.3028	0.9130	-0.01	0.03	0.03	0.00	-27.4156	9.2256				
		8	0.0017	-0.0001	44.99	-0.0003	-1.2359	1.0655	-0.0035	-0.0001	-0.0001	-0.0003	-0.2731	0.5371				
		9	0.0015	-0.0003	44.99	-0.0003	-1.2359	1.0655	-0.0035	-0.0001	-0.0001	-0.0003	-0.2731	0.5371				
87	1	7	0.898	-5.10	44.99	-0.0005	0.2246	0.6108	0.99	0.99	0.99	0.98	0.0938	0.7168				
		8	-0.0129	-0.0005	44.99	-0.0013	0.0674	0.6301	-0.0107	-0.0002	-0.0005	-0.0015	0.2497	0.7206				
		9	-0.0105	-0.0001	44.99	-0.0013	0.0674	0.6301	-0.0107	-0.0002	-0.0005	-0.0015	0.2497	0.7206				

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																										
87	2	0.899	0.0051	0.0068	0.0005	0.02	0.0001	0.0005	44.99	0.0008	0.0010	0.1316	0.4067	0.8377	0.7821	1.02	0.0046	0.0083	1.02	0.0005	0.0000	0.0005	0.0010	1.00	0.5520	0.0448	0.6624	0.6368
87	3	0.897	0.0138	0.0137	0.0007	1.02	0.0002	0.0007	44.99	0.0019	0.0019	0.0775	0.2600	0.7138	0.7138	1.03	0.0136	0.0162	1.03	0.0006	0.0002	0.0002	1.01	0.0018	0.2206	0.0727	0.6747	0.6859
87	4	0.897	0.0315	0.0311	0.0008	3.11	0.0003	0.0008	44.99	0.0044	0.0043	0.0525	0.1427	0.7002	0.7002	1.05	0.0283	0.0324	1.04	0.0007	0.0003	0.0003	1.02	0.0038	0.1328	0.0594	0.6758	0.6822
87	5	0.897	0.0499	0.0503	0.0014	6.22	0.0009	0.0014	44.99	0.0065	0.0064	0.0919	0.1448	0.6523	0.6428	1.08	0.0472	0.0515	1.07	0.0011	0.0008	0.0008	1.05	0.0066	0.1171	0.0810	0.6368	0.6481
87	6	0.897	0.0645	0.0667	0.0014	9.33	0.0010	0.0014	44.99	0.0081	0.0081	0.0842	0.1118	0.6290	0.6105	1.09	0.0636	0.0670	1.09	0.0013	0.0010	0.0010	1.07	0.0080	0.1045	0.0788	0.6204	0.6033
87	7	0.897	0.0768	0.0780	0.0015	12.45	0.0011	0.0015	44.99	0.0093	0.0092	0.0765	0.0961	0.6102	0.5894	1.11	0.0774	0.0802	1.11	0.0013	0.0010	0.0010	1.08	0.0093	0.0885	0.0676	0.6024	0.5806
87	8	0.898	0.0053	0.0068	0.0005	0.01	0.0001	0.0005	44.99	0.0008	0.0010	0.1302	0.4275	0.8321	0.7654	1.03	0.0050	0.0086	1.02	0.0005	0.0000	0.0000	1.01	0.0006	0.5008	0.0466	0.6385	0.6221
88	1	0.897	0.0008	0.0013	0.0002	-3.09	0.0002	0.0002	44.99	0.0000	0.0000	1.2969	0.9703	2.98	0.3566	0.0696	3.01	0.0006	0.0020	2.99	0.0002	0.0002	2.98	0.0001	-1.7885	-0.7385	1.1221	0.6204
88	2	0.897	0.0207	0.0194	0.0007	0.03	0.0002	0.0007	44.99	0.0029	0.0027	0.0564	0.2039	0.7008	0.7175	3.01	0.0182	0.0209	3.02	0.0006	0.0002	0.0002	3.00	0.0024	0.1692	0.0584	0.6730	0.6810
88	3	0.897	0.0305	0.0280	0.0008	1.06	0.0001	0.0008	44.99	0.0042	0.0039	0.0214	0.1502	0.7034	0.7046	3.01	0.0278	0.0291	3.03	0.0005	0.0002	0.0002	3.01	0.0037	0.1068	0.0511	0.6779	0.6817
88	4	0.897	0.0443	0.0421	0.0012	3.13	0.0005	0.0012	44.99	0.0060	0.0055	0.0635	0.1453	0.6850	0.6624	3.03	0.0400	0.0433	3.05	0.0008	0.0005	0.0005	3.02	0.0053	0.1071	0.0668	0.6679	0.6834

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																					
89	8	0.895	0.002	44.99	0.0336	5.99	6.03	6.03	6.04	0.0622	6.04	0.0631	0.0418	0.0002	0.0052	0.0336	0.7032	0.419	6.03	0.0006	0.0057	0.0622	0.6631
	9	0.0387	0.0010	0.0052	0.1363	0.6792	0.422	0.0004	0.0004	0.0576	0.0004	0.0057	0.0387	0.0010	0.0052	0.1363	0.6792	0.422	0.0004	0.0004	0.0057	0.0576	0.6631
90	1	0.896	-3.03	44.99	0.0276	9.04	9.05	9.05	9.03	0.0908	9.03	0.0834	0.0401	0.0002	0.0048	0.0276	0.7124	0.395	9.05	0.0007	0.0052	0.0908	0.6834
	9	0.0358	0.0007	0.0048	0.1102	0.6835	0.388	0.0002	0.0002	0.0339	0.0002	0.0052	0.0358	0.0007	0.0048	0.1102	0.6835	0.388	0.0002	0.0002	0.0052	0.0339	0.6834
90	2	0.896	0.09	44.99	0.0647	9.06	9.06	9.08	9.04	0.0923	9.04	0.6331	0.0553	0.0007	0.0070	0.0647	0.6540	0.562	9.06	0.0010	0.0072	0.0923	0.6331
	9	0.0543	0.0013	0.0070	0.1223	0.6452	0.570	0.0007	0.0007	0.0691	0.0007	0.6331	0.0543	0.0013	0.0070	0.1223	0.6452	0.570	0.0007	0.0007	0.0691	0.6331	
90	3	0.895	1.11	44.99	0.0639	9.07	9.07	9.08	9.05	0.0861	9.05	0.6244	0.0613	0.0007	0.0075	0.0639	0.6329	0.620	9.07	0.0010	0.0077	0.0861	0.6244
	9	0.0598	0.0013	0.0075	0.1163	0.6329	0.623	0.0008	0.0008	0.0696	0.0008	0.6244	0.0598	0.0013	0.0075	0.1163	0.6329	0.623	0.0008	0.0008	0.0696	0.6244	
90	4	0.895	5.17	44.99	0.0569	9.08	9.08	9.10	9.06	0.0748	9.06	0.6093	0.0726	0.0008	0.0087	0.0569	0.6234	0.720	9.08	0.0010	0.0088	0.0748	0.6093
	9	0.0711	0.0014	0.0087	0.1049	0.6133	0.713	0.0009	0.0009	0.0647	0.0009	0.6093	0.0711	0.0014	0.0087	0.1049	0.6133	0.713	0.0009	0.0009	0.0647	0.6093	
90	5	0.898	6.27	44.99	0.0494	9.10	9.09	9.11	9.07	0.0575	9.11	0.5884	0.0869	0.0008	0.0094	0.0494	0.5990	0.865	9.10	0.0009	0.0097	0.0575	0.5884
	9	0.0805	0.0014	0.0094	0.0901	0.5874	0.825	0.0010	0.0010	0.0625	0.0010	0.5884	0.0805	0.0008	0.0094	0.0901	0.5990	0.865	0.0009	0.0010	0.0625	0.5884	
90	6	0.896	9.37	44.99	0.0354	9.10	9.10	9.10	9.06	0.0463	9.10	0.5795	0.0933	0.0006	0.0109	0.0354	0.5858	0.930	9.10	0.0008	0.0107	0.0463	0.5795
	9	0.0869	0.0007	0.0109	0.0456	0.5832	0.895	0.0005	0.0005	0.0303	0.0005	0.5795	0.0869	0.0006	0.0109	0.0456	0.5832	0.895	0.0005	0.0005	0.0303	0.5795	
90	7	0.895	12.48	44.99	0.0106	9.10	9.09	9.10	9.06	0.0176	9.10	0.5751	0.1001	0.0002	0.0111	0.0106	0.5815	1.000	9.10	0.0003	0.0115	0.0176	0.5751
	9	0.0945	0.0004	0.0111	0.0254	0.5872	0.947	0.0002	0.0002	0.0124	0.0002	0.5751	0.0945	0.0002	0.0111	0.0254	0.5872	0.947	0.0002	0.0002	0.0124	0.5751	
90	8	0.896	0.06	44.99	0.0639	9.07	9.07	9.08	9.04	0.0886	9.08	0.6344	0.0554	0.0007	0.0069	0.0639	0.6469	0.565	9.08	0.0010	0.0072	0.0886	0.6344
	9	0.0540	0.0013	0.0069	0.1240	0.6404	0.572	0.0007	0.0007	0.0681	0.0007	0.6344	0.0540	0.0007	0.0069	0.1240	0.6404	0.572	0.0007	0.0007	0.0681	0.6344	
91	1	0.897	-3.01	44.99	0.0461	15.07	15.08	15.07	15.05	0.0719	15.07	0.6272	0.0703	0.0006	0.0082	0.0461	0.6332	0.707	15.08	0.0010	0.0088	0.0719	0.6272
	9	0.0654	0.0011	0.0082	0.0881	0.6303	0.695	0.0005	0.0005	0.0399	0.0005	0.6272	0.0654	0.0006	0.0082	0.0881	0.6303	0.695	0.0005	0.0005	0.0399	0.6272	
91	2	0.898	0.13	44.99	0.0425	15.08	15.09	15.09	15.07	0.0596	15.09	0.5963	0.0794	0.0007	0.0095	0.0425	0.6075	0.860	15.09	0.0010	0.0104	0.0596	0.5963
	9	0.0794	0.0013	0.0095	0.0824	0.6018	0.825	0.0007	0.0007	0.0465	0.0007	0.5963	0.0794	0.0013	0.0095	0.0824	0.6018	0.825	0.0007	0.0007	0.0465	0.5963	

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

← See Key →

Run	Pt.	Cd																		
91	3	7	0.898	1.13	44.99	15.09	15.09	15.09	15.09	15.09	15.09	15.09	15.09	15.09	15.09	15.09	15.09	15.09	15.09	15.09
		6	0.901	0.0007	0.0108	0.6021	0.0906	0.0008	0.0008	0.0008	0.0008	0.0008	0.0008	0.0008	0.0008	0.0008	0.0008	0.0008	0.0008	0.0008
		9	0.0822	0.0013	0.0097	0.5947	0.0852	0.0008	0.0008	0.0008	0.0008	0.0008	0.0008	0.0008	0.0008	0.0008	0.0008	0.0008	0.0008	0.0008
91	4	7	0.897	5.18	44.99	15.09	15.09	15.09	15.09	15.09	15.09	15.09	15.09	15.09	15.09	15.09	15.09	15.09	15.09	15.09
		6	0.957	0.0007	0.0113	0.5904	0.0961	0.0008	0.0008	0.0008	0.0008	0.0008	0.0008	0.0008	0.0008	0.0008	0.0008	0.0008	0.0008	0.0008
		9	0.0873	0.0010	0.0102	0.5883	0.0905	0.0006	0.0006	0.0006	0.0006	0.0006	0.0006	0.0006	0.0006	0.0006	0.0006	0.0006	0.0006	0.0006
91	5	7	0.896	6.29	44.99	15.09	15.09	15.09	15.09	15.09	15.09	15.09	15.09	15.09	15.09	15.09	15.09	15.09	15.09	15.09
		6	0.1033	0.0003	0.0120	0.5834	0.1033	0.0005	0.0005	0.0005	0.0005	0.0005	0.0005	0.0005	0.0005	0.0005	0.0005	0.0005	0.0005	0.0005
		9	0.0941	0.0005	0.0110	0.5864	0.0971	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002
91	6	7	0.897	9.39	44.99	15.09	15.09	15.09	15.09	15.09	15.09	15.09	15.09	15.09	15.09	15.09	15.09	15.09	15.09	15.09
		6	0.1044	0.0000	0.0121	0.5837	0.1063	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001
		9	0.1014	0.0003	0.0119	0.5877	0.1033	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
91	7	7	0.900	12.49	44.99	15.09	15.09	15.09	15.09	15.09	15.09	15.09	15.09	15.09	15.09	15.09	15.09	15.09	15.09	15.09
		6	0.1080	-0.0002	0.0125	0.5799	0.1134	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001
		9	0.1038	0.0004	0.0120	0.5815	0.1060	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000
91	8	7	0.897	0.10	44.99	15.09	15.09	15.09	15.09	15.09	15.09	15.09	15.09	15.09	15.09	15.09	15.09	15.09	15.09	15.09
		6	0.0851	0.0007	0.0103	0.6077	0.0864	0.0009	0.0009	0.0009	0.0009	0.0009	0.0009	0.0009	0.0009	0.0009	0.0009	0.0009	0.0009	0.0009
		9	0.0793	0.0012	0.0095	0.6047	0.0824	0.0007	0.0007	0.0007	0.0007	0.0007	0.0007	0.0007	0.0007	0.0007	0.0007	0.0007	0.0007	0.0007
92	6	7	0.600	-3.10	-0.00	0.03	-3.03	0.01	0.01	0.01	0.01	0.01	0.01	0.01	0.01	0.01	0.01	0.01	0.01	0.01
		6	0.0018	-0.0001	0.0002	0.7681	-0.0397	-0.0011	-0.0011	-0.0011	-0.0011	-0.0011	-0.0011	-0.0011	-0.0011	-0.0011	-0.0011	-0.0011	-0.0011	-0.0011
		9	0.0003	0.0001	0.0000	0.1295	-0.0381	-0.0014	-0.0014	-0.0014	-0.0014	-0.0014	-0.0014	-0.0014	-0.0014	-0.0014	-0.0014	-0.0014	-0.0014	-0.0014
92	7	7	0.599	-1.06	-0.00	0.04	-3.00	0.02	0.02	0.02	0.02	0.02	0.02	0.02	0.02	0.02	0.02	0.02	0.02	0.02
		6	0.0029	-0.0000	0.0004	0.6991	-0.0189	-0.0004	-0.0004	-0.0004	-0.0004	-0.0004	-0.0004	-0.0004	-0.0004	-0.0004	-0.0004	-0.0004	-0.0004	-0.0004
		9	0.0004	0.0002	0.0000	0.5550	-0.0190	-0.0008	-0.0008	-0.0008	-0.0008	-0.0008	-0.0008	-0.0008	-0.0008	-0.0008	-0.0008	-0.0008	-0.0008	-0.0008
92	8	7	0.600	-0.06	-0.00	0.05	-2.99	0.01	0.01	0.01	0.01	0.01	0.01	0.01	0.01	0.01	0.01	0.01	0.01	0.01
		6	0.0028	-0.0000	0.0003	0.7027	-0.0105	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001
		9	0.0003	0.0001	0.0000	0.8048	-0.0107	-0.0005	-0.0005	-0.0005	-0.0005	-0.0005	-0.0005	-0.0005	-0.0005	-0.0005	-0.0005	-0.0005	-0.0005	-0.0005
92	9	7	0.600	0.44	-0.00	0.04	-2.98	0.02	0.02	0.02	0.02	0.02	0.02	0.02	0.02	0.02	0.02	0.02	0.02	0.02
		6	0.0034	-0.0000	0.0005	0.7251	-0.0074	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000
		9	0.0009	0.0002	0.0000	0.4648	-0.0068	-0.0004	-0.0004	-0.0004	-0.0004	-0.0004	-0.0004	-0.0004	-0.0004	-0.0004	-0.0004	-0.0004	-0.0004	-0.0004
92	10	7	0.601	1.00	-0.00	0.05	-2.98	0.01	0.01	0.01	0.01	0.01	0.01	0.01	0.01	0.01	0.01	0.01	0.01	0.01
		6	0.0031	-0.0000	0.0004	0.6703	-0.0033	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
		9	0.0004	0.0001	0.0000	0.3832	-0.0025	-0.0002	-0.0002	-0.0002	-0.0002	-0.0002	-0.0002	-0.0002	-0.0002	-0.0002	-0.0002	-0.0002	-0.0002	-0.0002

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																				
92	11	7	0.599	5.08	-0.0000	-0.0003	-0.0411	0.04	-2.94	0.01	-2.98	0.2904	0.6523	0.0026	-0.0000	0.7027	0.0124	0.0007	0.0016	0.1164	0.6168	
		8	0.0017	-0.0000	0.0002	0.0003	15.6187	-2.2406	0.0108	0.0002	0.0013	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
		9	0.0015	0.0002	0.0001	0.0001	0.8443	0.5854	0.0396	0.0012	0.0051	0.1609	0.6451	0.0015	0.0002	0.5854	0.0396	0.0012	0.0051	0.1609	0.6451	0.6537
92	12	7	0.599	6.19	-0.0000	-0.0002	-0.2904	0.03	-2.90	0.02	-2.95	0.2138	0.6537	0.0017	-0.0000	0.6606	0.0376	0.0016	0.0049	0.1609	0.6451	0.6537
		8	0.0019	-0.0001	0.0002	0.0002	0.2632	0.6321	0.0631	0.0023	0.0077	0.1879	0.6096	0.0019	-0.0001	0.6321	0.0631	0.0023	0.0077	0.1879	0.6096	0.6096
		9	0.0011	0.0002	0.0001	0.0001	1.2392	0.8151	0.0646	0.0021	0.0077	0.1666	0.6005	0.0011	0.0002	0.8151	0.0646	0.0021	0.0077	0.1666	0.6005	0.6005
92	13	7	0.598	12.43	-0.0000	-0.0002	-0.2899	0.06	-2.95	0.02	-2.90	0.1580	0.5727	0.0017	-0.0000	0.6861	0.0792	0.0025	0.0090	0.1282	0.5652	0.5727
		8	0.0017	-0.0000	0.0002	0.0002	7.8756	-2.4189	0.0805	0.0020	0.0091	0.1580	0.5727	0.0017	-0.0000	0.6861	0.0792	0.0025	0.0090	0.1282	0.5652	0.5727
		9	0.0006	0.0002	0.0000	0.0000	1.8219	0.4931	-0.0105	-0.0005	-0.0014	0.2455	0.6917	0.0006	0.0002	0.4931	-0.0105	-0.0005	-0.0014	0.2455	0.6917	0.6917
93	1	7	0.600	-3.10	-0.0000	-0.0003	-0.1673	0.04	-1.03	0.01	-1.04	0.1432	0.6529	0.0023	-0.0000	0.7345	-0.0278	-0.0007	-0.0036	0.2063	0.6622	0.6529
		8	0.0035	-0.0000	0.0002	0.0003	9.3549	0.0651	-0.0274	-0.0011	-0.0036	0.1432	0.6529	0.0023	-0.0000	0.7345	-0.0278	-0.0007	-0.0036	0.2063	0.6622	0.6529
		9	0.0012	0.0002	-0.0000	0.0000	-9.3549	0.0651	-0.0274	-0.0011	-0.0036	0.1432	0.6529	0.0012	0.0002	0.0651	-0.0274	-0.0011	-0.0036	0.2063	0.6622	0.6529
93	2	7	0.599	-1.07	-0.0000	-0.0004	-0.0625	0.04	-1.00	0.02	-1.02	0.0807	0.6767	0.0035	-0.0000	0.6616	-0.0090	-0.0001	-0.0012	0.0807	0.6767	0.6767
		8	0.0035	-0.0000	0.0002	0.0004	0.9095	0.5482	-0.0082	-0.0004	-0.0011	0.0807	0.6767	0.0035	-0.0000	0.6616	-0.0090	-0.0001	-0.0012	0.0807	0.6767	0.6767
		9	0.0012	0.0002	-0.0000	0.0001	-9.3549	0.0651	-0.0274	-0.0011	-0.0036	0.1432	0.6529	0.0012	0.0002	0.0651	-0.0274	-0.0011	-0.0036	0.2063	0.6622	0.6529
93	3	7	0.600	-0.05	-0.0000	-0.0004	-0.0470	0.04	-0.98	0.02	-1.01	0.2138	0.6504	0.0009	-0.0000	0.6784	-0.0025	-0.0002	-0.0002	0.1500	0.7667	0.6504
		8	0.0035	-0.0000	0.0002	0.0004	1.1367	0.4648	-0.0017	-0.0002	-0.0002	0.2138	0.6504	0.0009	-0.0000	0.6784	-0.0025	-0.0002	-0.0002	0.1500	0.7667	0.6504
		9	0.0009	0.0002	0.0000	0.0000	-3.5123	0.6205	-0.0023	-0.0000	-0.0002	0.1500	0.7667	0.0009	0.0002	0.6205	-0.0023	-0.0000	-0.0002	0.1500	0.7667	0.6504
93	4	7	0.601	0.45	-0.0000	-0.0004	-0.0655	0.04	-0.97	0.01	-1.00	0.093	0.7941	0.0028	-0.0000	0.7041	0.0014	0.0003	0.0002	0.1500	0.7941	0.7941
		8	0.0028	-0.0000	0.0001	0.0004	3.5123	0.6205	-0.0023	-0.0000	-0.0002	0.093	0.7941	0.0028	-0.0000	0.7041	0.0014	0.0003	0.0002	0.1500	0.7941	0.7941
		9	0.0002	0.0001	0.0000	0.0000	-3.5123	0.6205	-0.0023	-0.0000	-0.0002	0.093	0.7941	0.0002	0.0001	0.6205	-0.0023	-0.0000	-0.0002	0.1500	0.7941	0.7941
93	5	7	0.600	1.02	-0.0000	-0.0004	-0.0476	0.05	-0.97	0.02	-1.00	0.4119	0.7158	0.0033	-0.0000	0.6700	0.0051	0.0004	0.0007	0.4119	0.7158	0.7158
		8	0.0033	-0.0000	0.0002	0.0004	1.2412	0.6825	-0.0061	-0.0000	-0.0007	0.4119	0.7158	0.0033	-0.0000	0.6700	0.0051	0.0004	0.0007	0.4119	0.7158	0.7158
		9	0.0009	0.0002	0.0000	0.0001	-1.2412	0.6825	0.0061	0.0000	0.0007	0.4119	0.7158	0.0009	0.0002	0.6825	0.0061	0.0000	0.0007	0.4119	0.7158	0.7158
93	6	7	0.599	5.10	-0.0000	-0.0003	-0.0558	0.04	-0.93	0.01	-0.97	0.275	0.6518	0.0031	-0.0000	0.6031	0.0250	0.0011	0.0032	0.1505	0.6518	0.6518
		8	0.0031	-0.0000	0.0002	0.0003	9.5384	-9.5384	0.0224	0.0006	0.0032	0.1505	0.6518	0.0031	-0.0000	0.6031	0.0250	0.0011	0.0032	0.1505	0.6518	0.6518
		9	0.0000	0.0002	-0.0000	0.0000	-9.5384	0.04	-0.93	0.01	-0.97	0.275	0.6518	0.0000	0.0002	0.04	-0.93	0.01	-0.97	0.275	0.6518	0.6518

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd																	
93	7	0.599	6.22	-0.002	0.0001	-0.0002	-0.1835	0.6060	0.03	-0.09	0.02	-0.93	0.2084	0.6338					
	8	0.0021	-0.0000	0.0002	0.0001	0.0002	1.3842	0.7221	0.03	0.0503	0.0021	0.0063	0.1574	0.6341					
	9	0.0008	0.0002	0.0001	0.0001	0.0001	0.0001	0.7221	0.03	0.0515	0.0016	0.0065							
93	7	0.599	9.29	0.002	0.0002	-0.0002	-0.3235	0.6899	0.03	-0.86	0.02	-0.90	0.1806	0.5969					
	8	0.0015	-0.0001	0.0002	0.0002	0.0002	1.2015	0.9606	0.03	0.0718	0.0025	0.0085	0.1565	0.5801					
	9	0.0011	0.0002	0.0002	0.0002	0.0002	0.0002	0.9606	0.03	0.0714	0.0022	0.0082							
93	7	0.598	12.44	-0.001	0.0001	-0.0001	-0.5628	0.7018	0.03	-0.86	0.02	-0.91	0.1235	0.5636					
	8	0.0010	-0.0001	0.0001	0.0001	0.0001	2.3432	1.0416	0.03	0.0841	0.0020	0.0094	0.0956	0.5610					
	9	0.0004	0.0002	0.0001	0.0001	0.0001	0.0001	1.0416	0.03	0.0848	0.0016	0.0095							
93	7	0.600	-0.03	0.004	0.0000	-0.0004	-0.0826	0.6502	0.04	-0.98	0.02	-1.01	-0.3727	0.7716					
	8	0.0031	-0.0000	0.0004	0.0004	0.0004	14.3138	3.8981	0.04	-0.0018	0.0001	-0.0002	2.0758	1.4801					
	9	0.0000	0.0002	0.0000	0.0000	0.0000	0.0000	3.8981	0.04	-0.0004	-0.0001	-0.0001							
94	1	0.599	3.08	-0.003	0.0003	-0.0003	-0.2079	0.6741	0.03	-0.03	0.02	-0.02	0.1378	0.6528					
	8	0.0022	-0.0000	0.0003	0.0003	0.0003	6.0938	1.2902	0.03	-0.0204	-0.0005	-0.0026	0.2280	0.6659					
	9	0.0001	0.0002	0.0000	0.0000	0.0000	0.0000	1.2902	0.03	-0.0195	-0.0008	-0.0025							
94	2	0.600	-1.03	0.004	0.0001	-0.0004	-0.0681	0.6751	0.04	-0.01	0.02	-0.00	-0.0182	0.6882					
	8	0.0033	-0.0000	0.0004	0.0004	0.0004	1.0368	0.5646	0.04	-0.0046	0.0000	-0.0004	0.5515	0.8155					
	9	0.0010	0.0002	0.0001	0.0001	0.0001	0.0001	0.5646	0.04	-0.0026	-0.0002	-0.0004							
94	3	0.600	-0.02	0.004	0.0000	-0.0004	-0.0480	0.6553	0.04	0.00	0.01	0.00	0.6502	0.7647					
	8	0.0034	-0.0000	0.0004	0.0004	0.0004	2.0346	0.5236	0.04	0.0024	0.0003	0.0003	0.5952	0.7194					
	9	0.0004	0.0002	0.0000	0.0000	0.0000	0.0000	0.5236	0.04	0.0034	-0.0000	0.0004	0.438	0.6234					
94	4	0.600	0.46	-0.005	0.0005	-0.0005	0.0191	0.7434	0.04	0.01	0.02	0.01	0.3952	0.7194					
	8	0.0035	0.0000	0.0005	0.0005	0.0005	1.6398	0.7066	0.04	0.0056	0.0004	0.0008	0.438	0.6234					
	9	0.0006	0.0002	0.0000	0.0000	0.0000	0.0000	0.7066	0.04	0.0060	0.0000	0.0007	0.438	0.6234					
94	5	0.601	1.02	0.004	0.0000	-0.0004	-0.0187	0.6493	0.04	0.02	0.01	0.01	0.2966	0.6912					
	8	0.0036	-0.0000	0.0004	0.0004	0.0004	1.7098	0.4770	0.04	0.0110	0.0006	0.0015	0.1195	0.6726					
	9	0.0005	0.0002	0.0000	0.0000	0.0000	0.0000	0.4770	0.04	0.0098	0.0002	0.0013	0.2966	0.6912					
94	6	0.601	3.07	-0.004	0.0004	-0.0004	-0.0177	0.6733	0.04	0.04	0.02	0.03	0.2170	0.6615					
	8	0.0035	-0.0000	0.0004	0.0004	0.0004	1.3496	0.7672	0.04	0.0294	0.0012	0.0038	0.1485	0.6589					
	9	0.0009	0.0002	0.0001	0.0001	0.0001	0.0001	0.7672	0.04	0.0275	0.0008	0.0036	0.2170	0.6615					
94	7	0.600	6.21	-0.001	0.0001	-0.0001	-0.3644	0.6037	0.03	0.08	0.01	0.07	0.2044	0.6311					
	8	0.0014	-0.0001	0.0001	0.0001	0.0001	1.1204	0.7260	0.03	0.0528	0.0021	0.0066	0.1639	0.6311					
	9	0.0012	0.0002	0.0000	0.0000	0.0000	0.0000	0.7260	0.03	0.0549	0.0018	0.0069	0.1639	0.6311					

See Key

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																		
94	8	0.599	9.30	0.002	-0.2044	0.003	0.11	0.01	0.09	0.1778	0.5894	0.0015	-0.0000	0.0002	0.9506	0.0744	0.0026	0.0087	0.1454	0.5766
	8	0.0010	0.0002	0.0002	1.2926	1.0518	0.0742	0.0021	0.0085	0.1778	0.5894	0.0010	0.0002	0.0002	0.9506	0.0742	0.0021	0.0085	0.1454	0.5766
94	9	0.599	12.43	-0.002	-0.3407	0.003	0.11	0.02	0.09	0.1066	0.5641	0.0015	-0.0001	0.0001	0.6883	0.0858	0.0018	0.0096	0.0817	0.5595
	8	0.0007	0.0002	0.0001	1.8610	1.1050	0.0862	0.0014	0.0096	0.1066	0.5641	0.0007	0.0002	0.0001	0.6883	0.0858	0.0018	0.0096	0.0817	0.5595
94	10	0.600	-0.02	-0.004	-0.0213	0.005	0.00	0.02	-0.00	0.7190	0.7217	0.0034	-0.0000	0.0004	0.6870	0.0020	0.0002	-0.0002	-0.1421	0.5056
	8	0.0006	0.0002	0.0001	1.8502	0.9521	0.0029	0.0000	0.0002	0.7190	0.7217	0.0006	0.0002	0.0001	0.6870	0.0020	0.0002	-0.0002	-0.1421	0.5056
95	1	0.599	-3.07	-0.003	-0.1538	0.003	0.97	0.02	0.97	0.1358	0.6584	0.0024	-0.0000	0.0003	0.7119	-0.0153	-0.0004	-0.0020	0.1358	0.6584
	8	0.0005	0.0002	0.0000	2.1898	0.7392	-0.0148	-0.0007	-0.0019	0.1358	0.6584	0.0005	0.0002	0.0000	0.7119	-0.0148	-0.0007	-0.0019	0.1358	0.6584
95	2	0.599	-1.06	-0.004	-0.0637	0.005	1.00	0.01	1.00	-1.7755	0.1376	0.0033	-0.0000	0.0004	0.6552	-0.0005	0.0001	-0.0000	-1.7755	0.1376
	8	0.0007	0.0002	0.0000	1.4629	0.6195	-0.0009	-0.0001	-0.0000	-1.7755	0.1376	0.0007	0.0002	0.0000	0.6552	-0.0009	-0.0001	-0.0000	-1.7755	0.1376
95	3	0.599	-0.02	-0.004	-0.0586	0.005	1.02	0.01	1.00	0.3359	0.7077	0.0031	-0.0001	0.0004	0.6841	0.0074	0.0005	0.0009	0.3359	0.7077
	8	0.0002	0.0001	-0.0000	-4.1979	0.6634	0.0075	0.0001	0.0009	0.3359	0.7077	0.0002	0.0001	-0.0000	0.6841	0.0074	0.0005	0.0009	0.3359	0.7077
95	4	0.599	0.46	-0.005	0.0092	0.004	1.03	0.01	1.01	0.2901	0.6933	0.0035	0.0000	0.0005	0.7183	0.0108	0.0006	0.0013	0.2901	0.6933
	8	0.0006	0.0002	0.0000	1.4869	0.5883	0.0104	0.0002	0.0013	0.2901	0.6933	0.0006	0.0002	0.0005	0.7183	0.0108	0.0006	0.0013	0.2901	0.6933
95	5	0.601	0.99	-0.005	0.0209	0.004	1.03	0.01	1.01	0.2539	0.6774	0.0038	0.0000	0.0005	0.6900	0.0160	0.0008	0.0021	0.2539	0.6774
	8	0.0010	0.0002	0.0000	1.0207	0.4718	0.0143	0.0003	0.0018	0.2539	0.6774	0.0010	0.0002	0.0005	0.6900	0.0160	0.0008	0.0021	0.2539	0.6774
95	6	0.600	3.11	-0.004	-0.0261	0.004	1.07	0.02	1.04	0.2081	0.6552	0.0032	-0.0000	0.0004	0.6647	0.0363	0.0015	0.0047	0.2081	0.6552
	8	0.0004	0.0002	0.0000	2.4717	0.9386	0.0341	0.0009	0.0045	0.2081	0.6552	0.0004	0.0002	0.0004	0.6647	0.0363	0.0015	0.0047	0.2081	0.6552
95	7	0.600	6.19	-0.002	-0.2576	0.004	1.10	0.02	1.08	0.1933	0.6211	0.0016	-0.0000	0.0002	0.7844	0.0577	0.0022	0.0071	0.1933	0.6211
	8	0.0015	0.0002	0.0002	0.9461	0.8640	0.0595	0.0019	0.0072	0.1933	0.6211	0.0015	0.0002	0.0002	0.7844	0.0577	0.0022	0.0071	0.1933	0.6211
95	8	0.599	9.30	-0.002	-0.3247	0.003	1.13	0.02	1.09	0.1698	0.5774	0.0014	-0.0000	0.0002	0.7564	0.0769	0.0026	0.0088	0.1698	0.5774
	8	0.0010	0.0002	0.0001	1.1615	0.8151	0.0779	0.0020	0.0088	0.1698	0.5774	0.0010	0.0002	0.0001	0.7564	0.0769	0.0026	0.0088	0.1698	0.5774

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key														
95	9	0.598	12.46	-0.0001	0.0001	-0.4600	0.003	0.7684	1.0372	0.0861	1.12	0.01	0.0016	0.0097	0.0984	0.5633
	8	0.0007	-0.0002	0.0001	0.0001	1.5937	0.0001	1.0372	0.0860	0.0860	0.0860	0.0012	0.0096	0.0721	0.5575	
95	10	0.600	-0.0000	0.0004	0.0000	-0.0860	0.005	0.7952	0.0070	1.01	0.02	0.0004	1.000	0.3290	0.6944	
	9	0.0025	0.0002	0.0000	0.0000	5.0021	0.0000	1.5025	0.0080	0.0080	0.0001	0.0010	0.0010	0.0814	0.6220	
96	1	0.59	-3.07	-0.00	0.0004	-0.0889	0.005	0.6943	2.99	2.99	0.01	0.0003	2.99	0.0548	0.6455	
	8	0.002	-0.0000	0.0001	0.0001	1.8896	0.0001	0.9133	-0.0044	-0.0044	-0.0000	-0.0006	-0.0006	0.4313	0.7185	
	9	0.0006	0.0002	0.0001	0.0001	1.8896	0.0001	0.9133	-0.0044	-0.0044	-0.0000	-0.0006	-0.0006	0.4313	0.7185	
96	2	0.600	-1.03	-0.00	0.0005	-0.0209	0.004	0.7292	0.0085	3.02	0.01	0.0005	3.02	0.3073	0.7319	
	8	0.0036	-0.0000	0.0001	0.0001	1.5870	0.0001	0.8426	0.0086	0.0086	0.0001	0.0011	0.0011	0.0895	0.6511	
	9	0.0008	0.0002	0.0001	0.0001	1.5870	0.0001	0.8426	0.0086	0.0086	0.0001	0.0011	0.0011	0.0895	0.6511	
96	3	0.600	-0.00	-0.00	0.0005	0.0094	0.005	0.6894	0.0170	3.03	0.02	0.0008	3.02	0.2376	0.6907	
	8	0.0037	0.0000	0.0000	0.0005	0.0094	0.0005	0.6894	0.0170	0.0170	0.0008	0.0023	0.0023	0.1146	0.6486	
	9	0.0009	0.0002	0.0001	0.0001	1.2616	0.0001	0.6538	0.0162	0.0162	0.0003	0.0021	0.0021	0.1146	0.6486	
96	4	0.600	0.48	-0.00	0.0004	-0.0860	0.004	0.7952	0.0225	3.04	0.01	0.0010	3.03	0.2247	0.6846	
	8	0.0025	-0.0000	0.0000	0.0004	-0.0860	0.0004	0.7952	0.0225	0.0225	0.0010	0.0030	0.0030	0.1286	0.6558	
	9	0.0004	0.0001	0.0000	0.0000	2.4512	0.0000	0.6638	0.0220	0.0220	0.0005	0.0028	0.0028	0.1286	0.6558	
96	5	0.601	1.03	-0.00	0.0005	0.0336	0.005	0.6995	0.0278	3.06	0.02	0.0011	3.04	0.2129	0.6754	
	8	0.0042	0.0000	0.0000	0.0005	0.0336	0.0005	0.6995	0.0278	0.0278	0.0011	0.0037	0.0037	0.1336	0.6644	
	9	0.0012	0.0002	0.0001	0.0001	1.0761	0.0001	0.7777	0.0253	0.0253	0.0006	0.0033	0.0033	0.1336	0.6644	
96	6	0.600	3.10	-0.00	0.0004	0.0190	0.004	0.6978	0.0459	3.08	0.02	0.0018	3.05	0.1995	0.6477	
	8	0.0034	0.0000	0.0000	0.0004	0.0190	0.0004	0.6978	0.0459	0.0459	0.0018	0.0057	0.0057	0.1419	0.6459	
	9	0.0011	0.0002	0.0001	0.0001	1.1341	0.0001	0.6588	0.0448	0.0448	0.0012	0.0057	0.0057	0.1419	0.6459	
96	7	0.600	6.21	-0.00	0.0002	-0.1643	0.004	0.6560	0.0687	3.12	0.02	0.0024	3.10	0.1769	0.6058	
	8	0.0020	-0.0000	0.0000	0.0002	-0.1643	0.0002	0.6560	0.0687	0.0687	0.0024	0.0082	0.0082	0.1564	0.5976	
	9	0.0001	0.0002	0.0000	0.0000	6.5252	0.0000	0.6245	0.0687	0.0687	0.0021	0.0082	0.0082	0.1564	0.5976	
96	8	0.600	9.30	-0.00	0.0002	-0.2129	0.003	0.6444	0.0824	3.13	0.02	0.0023	3.10	0.1412	0.5672	
	8	0.0019	-0.0000	0.0000	0.0002	-0.2129	0.0002	0.6444	0.0824	0.0824	0.0023	0.0092	0.0092	0.1126	0.5635	
	9	0.0013	0.0002	0.0002	0.0002	0.9189	0.0002	0.7853	0.0824	0.0824	0.0018	0.0092	0.0092	0.1126	0.5635	
96	9	0.599	12.47	-0.00	0.0002	-0.2109	0.004	0.7103	0.0827	3.11	0.02	0.0013	3.08	0.0816	0.5573	
	8	0.0019	-0.0000	0.0000	0.0002	-0.2109	0.0002	0.7103	0.0827	0.0827	0.0013	0.0091	0.0091	0.0552	0.5522	
	9	0.0008	0.0002	0.0000	0.0002	1.6092	0.0002	1.1895	0.0827	0.0827	0.0009	0.0091	0.0091	0.0552	0.5522	

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																					
96	10	7	0.599	-0.002	-0.000	0.0005	0.0015	0.04	3.04	0.01	3.01	0.2311	0.6792	0.0039	0.0000	0.0001	0.0015	0.6625	0.0179	0.0003	0.0024	0.1176	0.6480
97	3	7	0.598	-3.06	-0.0001	0.0002	-0.2516	0.03	5.99	0.02	6.00	0.3004	0.7632	0.0021	0.0001	0.0001	0.2932	0.6684	0.0073	0.0004	0.0011	0.0870	0.6894
97	4	7	0.599	-1.01	-0.0001	0.0004	-0.0582	0.05	6.02	0.02	6.02	0.2038	0.6854	0.0005	0.0000	0.0000	1.5235	0.1726	0.0241	0.0006	0.0032	0.1299	0.6810
97	5	7	0.600	0.000	-0.0001	0.0005	0.0037	0.05	6.04	0.02	6.03	0.1928	0.6720	0.0012	0.0002	0.0001	0.9933	0.6865	0.0355	0.0009	0.0047	0.1383	0.6748
97	6	7	0.599	0.51	0.0000	0.0005	0.0152	0.04	6.05	0.02	6.04	0.1908	0.6576	0.0008	0.0002	0.0000	1.2232	0.7554	0.0401	0.0015	0.0052	0.1340	0.6576
97	7	7	0.600	1.04	-0.0002	0.0004	-0.0371	0.05	6.05	0.02	6.05	0.1949	0.6575	0.0025	0.0000	0.0000	0.0371	0.8087	0.0463	0.0013	0.0060	0.1422	0.6497
97	8	7	0.600	-3.14	-0.0002	0.0003	-0.0867	0.07	6.07	0.02	6.07	0.1772	0.6220	0.0022	0.0000	0.0000	2.0648	0.7256	0.0614	0.0021	0.0074	0.1553	0.6220
97	9	7	0.599	6.24	-0.0002	0.0001	-0.0874	0.05	6.10	0.02	6.09	0.1585	0.5772	0.0012	0.0000	0.0000	1.0920	0.6574	0.0775	0.0020	0.0088	0.1329	0.5737
97	10	7	0.599	9.32	0.0000	0.0002	-0.1619	0.04	6.09	0.02	6.07	0.0940	0.5630	0.0020	0.0000	0.0000	0.8881	0.7419	0.0868	0.0016	0.0097	0.0740	0.5622
97	11	7	0.598	12.44	-0.0001	0.0002	-0.3830	0.04	6.08	0.02	6.05	0.0552	0.5570	0.0013	0.0000	0.0000	1.0420	0.7242	0.0824	0.0009	0.0089	0.0306	0.5551
97	12	7	0.599	0.02	-0.0000	0.0000	-0.0884	0.04	6.04	0.02	6.04	0.1913	0.6633	0.0024	0.0000	0.0000	8.0642	0.7857	0.0370	0.0014	0.0049	0.1324	0.6633
97	9	7	0.600	-0.0002	0.0000	0.0000	-0.0642	1.8114	1.8114	0.0009	0.0048	0.1324	0.6633	0.0001	0.0000	0.0000	0.0642	1.8114	0.0362	0.0009	0.0048	0.1324	0.6633

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key															
98	1	7	0.599	-0.0004	-0.0725	0.05	9.02	0.009	9.01	0.1939	0.6885						
		8	0.0025	0.0000	0.2932	0.8026	0.0244	0.0005	0.033	0.1163	0.6756						
		9	0.0003	0.0002		0.4213	0.0242	0.0005	0.0032								
98	2	7	0.600	-0.0003	-0.0941	0.05	9.05	0.02	9.04	0.1862	0.6578						
		8	0.0025	0.0000	0.135	0.782	0.0436	0.0016	0.057	0.1336	0.6587						
		9	0.0000	0.0002	0.8602	0.6615	0.0443	0.0011	0.0058								
98	3	7	0.600	-0.0004	-0.0131	0.05	9.07	0.02	9.05	0.1871	0.6395						
		8	0.0036	0.0000	0.4974	0.741	0.0521	0.0019	0.066	0.1412	0.6469						
		9	0.0002	0.0002		1.3484	0.0521	0.0014	0.0067								
98	4	7	0.599	-0.0004	-0.0544	0.05	9.07	0.02	9.06	0.1860	0.6367						
		8	0.0031	0.0000	0.3369	0.6439	0.0547	0.0020	0.069	0.1450	0.6329						
		9	0.0003	0.0002		0.6184	0.0551	0.0015	0.0069								
98	5	7	0.600	-0.0005	0.0258	0.04	9.08	0.02	9.07	0.1749	0.6278						
		8	0.0039	0.0000	0.259	0.919	0.0602	0.0021	0.075	0.1477	0.6245						
		9	0.0006	0.0002		1.1352	0.0588	0.0017	0.0073								
98	6	7	0.600	-0.0003	-0.0419	0.05	9.10	0.02	9.08	0.1647	0.5983						
		8	0.0028	0.0000	0.8179	0.806	0.0753	0.0024	0.090	0.1447	0.5953						
		9	-0.0002	0.0002		-1.5845	0.0729	0.0021	0.0086								
98	7	7	0.600	-0.0003	-0.0522	0.05	9.11	0.02	9.08	0.1209	0.5661						
		8	0.0021	0.0000	0.8418	0.402	0.0839	0.0020	0.095	0.1011	0.5629						
		9	0.0020	0.0003		0.7879	0.0837	0.0016	0.0094								
98	8	7	0.600	-0.0001	-0.5022	0.04	9.09	0.02	9.06	0.0725	0.5579						
		8	0.0009	0.0000	0.5022	0.9569	0.0836	0.0012	0.093	0.0496	0.5546						
		9	0.0012	0.0003		0.9323	0.0841	0.0008	0.0093								
98	9	7	0.599	-0.0007	-0.6326	0.03	9.08	0.02	9.05	0.0429	0.5613						
		8	0.0007	0.0000	0.7643	0.9985	0.0845	0.0007	0.094	0.0219	0.5615						
		9	0.0002	0.0002		1.3279	0.0825	0.0003	0.0092								
98	10	7	0.600	-0.0004	-0.0818	0.05	9.07	0.02	9.06	0.1841	0.6418						
		8	0.0024	0.0000	0.3024	0.7502	0.0505	0.0018	0.064	0.1442	0.6490						
		9	0.0011	0.0002		0.7338	0.0520	0.0015	0.0067								
99	1	7	0.599	-0.0004	-0.0582	0.05	12.07	0.02	12.04	0.1770	0.6648						
		8	0.0033	0.0000	0.3800	0.6661	0.0413	0.0014	0.055	0.1268	0.6638						
		9	0.0009	0.0002		0.7276	0.0425	0.0010	0.0056								

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key													
99	2	7	0.600	-1.000	-0.000	-0.000	-0.1013	0.7959	0.04	12.10	0.02	12.06	0.1720	0.6311	
		8	0.0024	-0.0000	0.0000	-0.0000	-3.5018	0.0229	0.582	0.0020	0.0073	0.1428	0.6273		
		9	-0.0003	0.0002	-0.0000	-0.0000	0.0229	0.0016	0.582	0.0016	0.0073	0.1428	0.6273		
99	3	7	0.600	0.05	0.0004	0.0004	-0.0509	0.8013	0.05	12.11	0.02	12.07	0.1603	0.6214	
		8	0.0025	-0.0000	0.0000	0.0000	-2.0282	0.8706	0.021	0.0018	0.0081	0.1403	0.6131		
		9	0.0006	0.0002	0.0001	0.0001	0.8706	0.0019	0.02	0.0018	0.0081	0.1403	0.6131		
99	4	7	0.600	0.53	0.0004	0.0004	-0.0273	0.6610	0.05	12.12	0.02	12.07	0.1601	0.6162	
		8	0.0034	-0.0000	0.0000	0.0000	1.6127	0.8349	0.022	0.0019	0.0085	0.1403	0.6105		
		9	0.0007	0.0002	0.0001	0.0001	0.8349	0.0019	0.02	0.0019	0.0085	0.1403	0.6105		
99	5	7	0.600	1.08	0.0005	0.0005	0.0342	0.7481	0.05	12.12	0.02	12.07	0.1606	0.6090	
		8	0.0035	0.0000	0.0000	0.0000	1.3800	0.7276	0.023	0.0019	0.0086	0.1392	0.6002		
		9	0.0009	0.0002	0.0001	0.0001	0.7276	0.0019	0.02	0.0019	0.0086	0.1392	0.6002		
99	6	7	0.600	3.16	0.0003	0.0003	-0.0378	0.7901	0.04	12.13	0.02	12.08	0.1445	0.5763	
		8	0.0023	-0.0000	0.0000	0.0000	1.8405	0.8398	0.023	0.0019	0.0091	0.1217	0.5723		
		9	0.0007	0.0002	0.0001	0.0001	0.8398	0.0019	0.02	0.0019	0.0091	0.1217	0.5723		
99	7	7	0.600	6.25	0.0002	0.0002	-0.3086	0.7682	0.03	12.12	0.02	12.07	0.0836	0.5655	
		8	0.0014	-0.0000	0.0000	0.0000	1.5253	0.7833	0.014	0.0011	0.0099	0.0631	0.5613		
		9	0.0008	0.0002	0.0001	0.0001	0.7833	0.0011	0.02	0.0011	0.0099	0.0631	0.5613		
99	8	7	0.600	9.29	0.0002	0.0002	-0.3468	0.7682	0.03	12.10	0.02	12.05	0.0524	0.5600	
		8	0.0013	-0.0000	0.0000	0.0000	1.5998	1.0307	0.008	0.0005	0.0094	0.0304	0.5590		
		9	0.0009	0.0002	0.0001	0.0001	1.0307	0.0005	0.02	0.0005	0.0092	0.0304	0.5590		
99	9	7	0.599	12.47	0.0001	0.0001	-0.3625	0.7559	0.03	12.10	0.02	12.05	0.0346	0.5619	
		8	0.0014	-0.0001	0.0000	0.0000	2.3288	1.1185	0.006	0.0002	0.0097	0.0169	0.5607		
		9	0.0005	0.0002	0.0001	0.0001	1.1185	0.0002	0.02	0.0002	0.0097	0.0169	0.5607		
99	10	7	0.600	0.02	0.0005	0.0005	0.0174	0.6657	0.04	12.11	0.02	12.06	0.1587	0.6209	
		8	0.0040	0.0000	0.0000	0.0000	1.7668	0.8837	0.020	0.0018	0.0079	0.1408	0.6158		
		9	0.0008	0.0002	0.0001	0.0001	0.8837	0.0018	0.02	0.0018	0.0079	0.1408	0.6158		
100	1	7	0.598	-2.98	0.0004	0.0004	-0.0774	0.6991	0.04	15.08	0.02	15.07	0.1690	0.6400	
		8	0.0029	-0.0000	0.0000	0.0000	5.4853	1.8572	0.019	0.0015	0.0072	0.1363	0.6239		
		9	0.0002	0.0002	0.0000	0.0000	1.8572	0.0015	0.02	0.0015	0.0072	0.1363	0.6239		
100	2	7	0.599	-0.97	0.0003	0.0003	-0.0736	0.6578	0.04	15.10	0.02	15.08	0.1541	0.6134	
		8	0.0028	-0.0000	0.0000	0.0000	-11.4252	-1.4067	0.022	0.0019	0.0086	0.1359	0.6007		
		9	-0.0001	0.0002	0.0000	0.0000	-1.4067	0.0019	0.02	0.0019	0.0086	0.1359	0.6007		

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

See Key →

Run	Pt.	Cd																					
100	5	7 6 9	0.600 0.0032 0.0002	0.04 -0.0000 0.0002	-0.00 0.0000 0.0000	0.04 0.0004 0.0000	-0.004 0.0000 0.0000	0.0587 4.8876	0.05 0.6681 1.1861	15.11 0.0772 0.0769	0.02 0.0024 0.0020	15.09 0.0092 0.0089	0.1562 0.1334	0.5970 0.5839									
100	4	7 6 9	0.600 0.0030 0.0008	0.54 -0.0000 0.0002	-0.00 0.0004 0.0001	0.04 0.6869 0.6039	-0.004 0.0000 0.0000	-0.0591 1.5684	0.04 0.6869 0.6039	15.11 0.0788 0.0782	0.02 0.0024 0.0020	15.09 0.0093 0.0090	0.1537 0.1283	0.5902 0.5782									
100	5	7 6 9	0.600 0.0031 0.0002	1.09 -0.0000 0.0002	-0.00 0.0004 0.0000	0.04 0.7039 -0.8841	-0.004 0.0000 0.0000	-0.0354 -5.0587	0.04 0.7039 -0.8841	15.11 0.0809 0.0802	0.02 0.0023 0.0018	15.09 0.0093 0.0091	0.1425 0.1172	0.5751 0.5719									
100	6	7 6 9	0.600 0.0022 0.0006	3.16 -0.0000 0.0002	-0.00 0.0003 0.0001	0.04 0.7717 0.7907	-0.003 0.0000 0.0000	-0.0702 2.1122	0.04 0.7717 0.7907	15.10 0.0866 0.0863	0.02 0.0017 0.0013	15.08 0.0098 0.0097	0.0991 0.0808	0.5692 0.5624									
100	7	7 6 9	0.599 0.0017 0.0003	6.23 -0.0000 0.0002	-0.00 0.0002 0.0000	0.03 0.7427 1.2602	-0.002 0.0000 0.0000	-0.1722 3.9872	0.03 0.7427 1.2602	15.09 0.0837 0.0854	0.02 0.0010 0.0007	15.07 0.0094 0.0094	0.0610 0.0431	0.5616 0.5554									
100	8	7 6 9	0.599 0.0014 0.0009	9.31 -0.0000 0.0002	-0.00 0.0003 0.0001	0.03 1.0642 0.8950	-0.003 0.0000 0.0000	-0.1983 1.5133	0.03 1.0642 0.8950	15.09 0.0863 0.0852	0.02 0.0007 0.0003	15.06 0.0097 0.0095	0.0410 0.0189	0.5624 0.5604									
100	9	7 6 9	0.599 0.0013 0.0006	12.50 -0.0000 0.0002	-0.00 0.0002 0.0001	0.03 0.8001 0.7871	-0.002 0.0000 0.0000	-0.3464 1.8305	0.03 0.8001 0.7871	15.09 0.0915 0.0916	0.02 0.0005 0.0002	15.06 0.0102 0.0102	0.0319 0.0119	0.5612 0.5607									
100	10	7 6 9	0.600 0.0028 0.0000	0.03 -0.0000 0.0002	-0.00 0.0004 0.0000	0.04 0.7187 6.6228	0.004 0.0000 0.0000	-0.0714 20.5765	0.04 0.7187 6.6228	15.11 0.0775 0.0771	0.02 0.0024 0.0020	15.09 0.0092 0.0089	0.1555 0.1327	0.5974 0.5829									
101	1	7 6 9	0.599 0.0100 0.0131	-3.07 -0.0005 0.0007	-0.00 0.0012 0.0018	-2.96 0.6421 0.6861	-0.002 0.0018 0.0018	0.2903 0.2852	-2.96 0.6421 0.6861	2.98 -0.0069 -0.0372	3.03 -0.0001 -0.0014	-3.06 -0.0009 -0.0049	0.1089 0.1959	0.6869 0.6675									
101	2	7 6 9	0.600 0.0103 0.0122	-1.06 -0.0005 0.0007	-0.00 0.0013 0.0017	-2.96 0.6335 0.7215	-0.003 0.0013 0.0017	0.2836 0.3046	-2.96 0.6335 0.7215	3.00 0.0078 -0.0183	3.02 0.0004 -0.0008	-3.04 -0.0025	0.2730 0.2262	0.6849 0.6904									
101	3	7 6 9	0.600 0.0092 0.0137	-0.003 -0.0005 0.0007	-0.00 0.0011 0.0018	-2.96 0.6340 0.6908	-0.003 0.0011 0.0018	0.2825 0.2851	-2.96 0.6340 0.6908	3.02 0.0167 -0.0096	3.03 0.0007 -0.0005	-3.02 0.0022 -0.0014	0.2218 0.2770	0.6578 0.7327									

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd																			
101	4	7	0.600	0.47	-0.00	0.2806	-2.96	3.07	3.02	-3.02	0.2143	0.6640									
		6	-0.0097	-0.0005	-0.0013	0.2952	0.6724	0.0207	0.0008	0.0027	0.3067	0.7474									
		9	0.0128	0.0007	0.0017		0.6895	-0.0068	-0.0004	-0.0010											
101	5	7	0.600	1.01	0.00	0.2898	-2.96	3.04	3.03	-3.01	0.1993	0.6547									
		6	-0.0083	-0.0004	-0.0010	0.2873	0.6496	0.0272	0.0010	0.0035	0.6050	0.9282									
		9	0.0137	0.0007	0.0018		0.6849	-0.0021	-0.0002	-0.0004											
101	6	7	0.600	3.14	0.00	0.2725	-2.96	3.07	3.03	-2.99	0.1962	0.6388									
		6	-0.0092	-0.0005	-0.0012	0.3012	0.6591	0.0454	0.0017	0.0058	0.1251	0.6039									
		9	0.0130	0.0007	0.0018		0.7001	0.0124	0.0003	0.0015											
101	7	7	0.600	6.20	0.00	0.2541	-2.96	3.10	3.03	-2.95	0.1710	0.6047									
		6	-0.0113	-0.0005	-0.0014	0.2990	0.6430	0.0688	0.0023	0.0083	0.1574	0.6339									
		9	0.0130	0.0007	0.0017		0.6891	0.0413	0.0013	0.0052											
101	8	7	0.599	9.26	-0.00	0.2424	-2.96	3.11	3.02	-2.92	0.1317	0.5615									
		6	-0.0120	-0.0005	-0.0015	0.3283	0.6458	0.0822	0.0021	0.0092	0.1635	0.5950									
		9	0.0118	0.0007	0.0017		0.7449	0.0657	0.0021	0.0078											
101	9	7	0.599	12.45	-0.00	0.2426	-2.96	3.09	3.03	-2.91	0.0797	0.5530									
		6	-0.0116	-0.0005	-0.0014	0.3008	0.6189	0.0832	0.0013	0.0092	0.1232	0.5618									
		9	0.0133	0.0008	0.0019		0.7265	0.0817	0.0020	0.0091											
101	10	7	0.600	-0.02	-0.00	0.2919	-2.96	3.01	3.02	-3.02	0.2253	0.6641									
		6	-0.0092	-0.0005	-0.0012	0.2954	0.6510	0.0168	0.0007	0.0022	0.2660	0.7262									
		9	0.0128	0.0007	0.0017		0.6856	-0.0102	-0.0005	-0.0014											
102	1	7	0.598	3.09	-0.00	0.0272	0.53	-0.53	-0.49	0.47	0.1645	0.6826									
		6	0.0046	0.0000	0.0006	-0.4897	0.7095	-0.0239	-0.0007	-0.0032	0.2361	0.6804									
		9	-0.0017	0.0001	-0.0002		0.7064	-0.0192	-0.0009	-0.0026											
102	2	7	0.600	-1.05	-0.00	0.0709	0.53	-0.51	-0.49	0.50	0.1119	0.7431									
		6	0.0056	0.0000	0.0008	-0.9735	0.7075	-0.0069	-0.0001	-0.0010	0.5913	0.8472									
		9	-0.0010	0.0001	-0.0001		0.9060	-0.0023	-0.0002	-0.0004											
102	3	7	0.600	-0.01	-0.09	0.0710	0.53	-0.50	-0.49	0.51	0.8609	0.1983									
		6	0.0055	0.0000	0.0007	-0.8718	0.7170	0.0009	0.0001	0.0000	0.0301	0.5922									
		9	-0.0009	0.0001	-0.0002		1.1166	0.0055	0.0000	0.0006											
102	4	7	0.600	0.47	-0.00	0.0590	0.53	-0.49	-0.49	0.52	0.3728	0.6143									
		6	0.0051	0.0000	0.0007	-0.6344	0.7113	0.0039	0.0001	0.0008	0.0853	0.6045									
		9	-0.0013	0.0001	-0.0002		0.9678	0.0073	0.0001	0.0008											

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd																			
102	5	7	0.600	0.99	-0.00	0.53	-0.48	0.52	0.2713	0.6336	0.0046	0.0000	0.0000	0.0000	0.0498	0.6980	0.0079	0.0010	0.1238	0.6422
		8	0.0046	0.0000	0.0006	0.6980	0.0079	0.0010			-0.0012	0.0001	-0.0002	-0.7832	0.9610	0.0117	0.0002	0.0015		
		9	-0.0012	0.0001	-0.0002	-0.7832	0.0117	0.0002			0.600	0.0000	0.0000	0.0379	0.6301	0.0049	0.0004	0.0015		
102	6	7	0.600	5.10	-0.00	0.53	-0.45	0.54	0.2043	0.6387	0.0049	0.0000	0.0006	0.5079	0.8672	0.0049	0.0035	0.1422	0.6498	
		8	0.0049	0.0000	0.0006	0.8672	0.0295	0.0035			-0.0017	0.0001	-0.0002	-0.5079	0.6301	0.0011	0.0035	0.2043	0.6387	
		9	-0.0017	0.0001	-0.0002	-0.5079	0.0295	0.0035			0.601	0.0000	0.0000	0.0481	0.7115	0.0049	0.0008	0.0035	0.1422	0.6498
102	7	7	0.601	6.21	-0.00	0.53	-0.39	0.58	0.1991	0.6197	0.0043	0.0000	0.0006	0.7861	0.9201	0.0021	0.0071	0.1623	0.6176	
		8	0.0043	0.0000	0.0006	0.9201	0.0542	0.0071			-0.0012	0.0001	-0.0002	-0.7861	0.7115	0.0021	0.0071	0.1623	0.6197	
		9	-0.0012	0.0001	-0.0002	-0.7861	0.0542	0.0071			0.600	0.0000	0.0000	0.0526	0.6823	0.0025	0.0067	0.1726	0.5865	
102	8	7	0.600	9.33	0.00	0.53	-0.36	0.60	0.1726	0.5865	0.0032	0.0000	0.0004	-1.6493	0.8081	0.0020	0.0067	0.1359	0.5691	
		8	0.0032	0.0000	0.0004	0.8081	0.0768	0.0067			-0.0007	0.0002	-0.0001	-1.6493	0.6823	0.0025	0.0067	0.1726	0.5865	
		9	-0.0007	0.0002	-0.0001	-1.6493	0.0768	0.0067			0.600	0.0000	0.0000	0.0698	0.6510	0.0018	0.0096	0.1111	0.5583	
102	9	7	0.600	12.45	0.00	0.52	-0.37	0.59	0.1111	0.5583	0.0032	0.0000	0.0002	-0.6913	0.7277	0.0018	0.0096	0.0769	0.5560	
		8	0.0032	0.0000	0.0002	0.7277	0.0863	0.0096			-0.0014	0.0001	-0.0002	-0.6913	0.6510	0.0018	0.0096	0.1111	0.5583	
		9	-0.0014	0.0001	-0.0002	-0.6913	0.0863	0.0096			0.599	0.0000	0.0000	0.0287	0.6685	0.0045	0.0050	0.1960	0.8991	
102	10	7	0.599	-0.02	-0.00	0.53	-0.50	0.50	0.1960	0.8991	0.0051	0.0000	0.0006	0.0656	0.7823	0.0045	0.0050	0.0006	0.5701	
		8	0.0051	0.0000	0.0006	0.7823	0.0045	0.0050			-0.0015	0.0002	-0.0002	-0.6656	0.6685	0.0045	0.0050	0.1593	0.6728	
		9	-0.0015	0.0002	-0.0002	-0.6656	0.0045	0.0050			0.600	0.0001	0.0009	0.0789	0.9989	0.0002	0.0000	0.2379	0.6645	
103	1	7	0.600	-3.06	-0.00	1.03	-1.02	1.01	0.1593	0.6728	0.0057	0.0001	0.0009	0.1699	0.7952	0.0007	0.0021	0.0430	0.7011	
		8	0.0057	0.0001	0.0009	0.7952	0.0007	0.0021			-0.0032	0.0001	-0.0005	-0.1699	0.6989	0.0007	0.0021	0.1593	0.6728	
		9	-0.0032	0.0001	-0.0005	-0.1699	0.0007	0.0021			0.601	0.0001	0.0009	0.0823	0.6851	0.0001	0.0003	0.1038	0.7011	
103	2	7	0.601	-1.06	-0.00	1.03	-0.99	1.03	0.1038	0.7011	0.0070	0.0001	0.0009	0.1767	0.8463	0.0001	0.0003	0.0430	0.2756	
		8	0.0070	0.0001	0.0009	0.8463	0.0001	0.0003			-0.0029	0.0001	-0.0005	-0.1767	0.6851	0.0001	0.0003	0.1038	0.7011	
		9	-0.0029	0.0001	-0.0005	-0.1767	0.0001	0.0003			0.601	0.0001	0.0009	0.0908	0.6732	0.0000	0.0000	0.0386	0.8710	
103	3	7	0.601	-0.05	-0.00	1.03	-0.98	1.04	0.0386	0.8710	0.0076	0.0001	0.0010	0.1712	0.7874	0.0000	0.0007	0.1470	0.5972	
		8	0.0076	0.0001	0.0010	0.7874	0.0000	0.0007			-0.0031	0.0001	-0.0004	-0.1712	0.6732	0.0000	0.0007	0.0386	0.8710	
		9	-0.0031	0.0001	-0.0004	-0.1712	0.0000	0.0007			0.601	0.0001	0.0010	0.0839	0.6767	0.0002	0.0002	0.0386	0.5972	
103	4	7	0.601	0.47	-0.00	1.03	-0.97	1.05	0.7133	0.6305	0.0076	0.0001	0.0010	0.1623	0.8831	0.0002	0.0002	0.1071	0.6242	
		8	0.0076	0.0001	0.0010	0.8831	0.0002	0.0002			-0.0030	0.0000	-0.0005	-0.1623	0.6767	0.0002	0.0002	0.0713	0.6305	
		9	-0.0030	0.0000	-0.0005	-0.1623	0.0002	0.0002			0.600	0.0001	0.0009	0.0850	0.6616	0.0003	0.0003	0.3232	0.6391	
103	5	7	0.600	0.99	-0.00	1.03	-0.97	1.05	0.3232	0.6391	0.0071	0.0001	0.0009	0.1309	0.9177	0.0003	0.0003	0.1258	0.6459	
		8	0.0071	0.0001	0.0009	0.9177	0.0003	0.0003			-0.0030	0.0000	-0.0005	-0.1309	0.6616	0.0003	0.0003	0.3232	0.6391	
		9	-0.0030	0.0000	-0.0005	-0.1309	0.0003	0.0003			0.600	0.0001	0.0009	0.0850	0.6616	0.0003	0.0003	0.1258	0.6459	

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd																		
103	6	7	0.601	5.14	-0.00	0.0989	1.03	-0.74	-1.00	1.09	0.2166	0.6464							
		8	0.0064	0.0001	0.0008	0.1219	0.6870	0.0248	0.0010	0.0032	0.1483	0.6510							
		9	-0.0035	0.0000	-0.0006	-0.1219	0.8578	0.0333	0.0009	0.0043									
103	7	7	0.601	6.24	-0.00	0.1025	1.03	-0.89	-1.00	1.12	0.2022	0.6231							
		8	0.0066	0.0001	0.0009	0.3184	0.6869	0.0519	0.0021	0.0064	0.1606	0.6081							
		9	-0.0026	0.0001	-0.0004	-0.3184	0.7740	0.0602	0.0019	0.0073									
103	8	7	0.601	9.34	0.00	0.0633	1.02	-0.86	-1.00	1.13	0.1760	0.5914							
		8	0.0055	0.0000	0.0007	0.3679	0.7064	0.0729	0.0025	0.0066	0.1325	0.5670							
		9	-0.0024	0.0001	-0.0003	-0.3679	0.6986	0.0781	0.0020	0.0088									
103	9	7	0.599	12.47	0.00	0.0517	1.02	-0.86	-0.99	1.12	0.1173	0.5593							
		8	0.0058	0.0000	0.0007	0.3603	0.6853	0.0843	0.0019	0.0094	0.0733	0.5532							
		9	-0.0023	0.0001	-0.0003	-0.3603	0.7855	0.0659	0.0012	0.0095									
103	10	7	0.600	-0.02	-0.00	0.0841	1.03	-0.98	-1.00	1.05	-0.2136	0.9775							
		8	0.0068	0.0001	0.0009	0.1659	0.6808	-0.0017	0.0000	-0.0003	-0.0662	0.5089							
		9	-0.0033	0.0001	-0.0005	-0.1659	0.3022	0.0073	0.0000	0.0008									
104	1	7	0.599	-3.10	-0.00	0.1092	2.00	-2.04	-2.00	1.97	0.1554	0.6622							
		8	0.0097	0.0002	0.0013	0.0286	0.6855	-0.0326	-0.0010	-0.0043	0.2641	0.6741							
		9	-0.0074	-0.0000	-0.0010	-0.0286	0.7216	-0.0105	-0.0005	-0.0014									
104	2	7	0.600	-1.05	-0.00	0.1104	2.00	-2.02	-2.00	1.99	0.1412	0.6992							
		8	0.0112	0.0002	0.0014	0.0619	0.6510	-0.0127	-0.0003	-0.0017	-0.0016	0.5839							
		9	-0.0068	-0.0000	-0.0011	-0.0619	0.8063	0.0046	-0.0000	-0.0005									
104	3	7	0.600	-0.03	-0.00	0.1145	2.00	-1.99	-2.00	2.00	0.0747	0.7536							
		8	0.0114	0.0002	0.0014	0.0587	0.6522	-0.0057	-0.0002	-0.0008	0.1141	0.6313							
		9	-0.0065	-0.0000	-0.0010	-0.0587	0.7977	0.0115	0.0002	0.0014									
104	4	7	0.599	0.46	-0.00	0.1011	2.00	-1.99	-2.00	2.01	-0.0950	0.9076							
		8	0.0106	0.0002	0.0013	0.0504	0.6404	-0.0023	0.0000	-0.0004	-0.1260	0.6318							
		9	-0.0077	-0.0000	-0.0011	-0.0504	0.7257	0.0146	0.0003	0.0018									
104	5	7	0.600	1.01	-0.00	0.1222	2.01	-1.99	-2.00	2.02	0.6318	0.4602							
		8	0.0115	0.0002	0.0014	0.0384	0.6471	-0.0018	0.0002	0.0001	0.1261	0.6285							
		9	-0.0068	-0.0000	-0.0010	-0.0384	0.7924	0.0192	0.0004	0.0024									
104	6	7	0.600	3.07	-0.00	0.1250	2.00	-1.95	-2.00	2.04	0.2293	0.6309							
		8	0.0108	0.0002	0.0014	0.0426	0.6749	-0.0194	0.0008	0.0024	0.1479	0.6468							
		9	-0.0072	-0.0000	-0.0011	-0.0426	0.7660	0.0406	0.0012	0.0052									

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Pt. Cd	See Key										
104	7	0.600	6.18	-0.000	-0.000	0.1131	2.00	-1.92	-2.00	2.08	0.1991	0.6320
	8	0.0095	0.0002	0.0012	0.0012	0.0118	0.6695	0.0448	0.0017	0.0056	0.1588	0.6042
	9	-0.0062	-0.0000	-0.0009	-0.0009	0.0118	0.8003	0.0631	0.0020	0.0076		
104	7	0.600	9.30	0.000	0.000	0.0985	2.00	-1.88	-1.99	2.09	0.1793	0.5983
	8	0.0095	0.0001	0.0012	0.0012	0.0230	0.6565	0.0690	0.0024	0.0082	0.1246	0.5633
	9	-0.0061	-0.0000	-0.0009	-0.0009	0.0230	0.7989	0.0803	0.0020	0.0090		
104	7	0.598	12.46	0.000	0.000	0.0933	2.00	-1.87	-1.99	2.07	0.1304	0.5609
	8	0.0098	0.0001	0.0013	0.0013	0.0215	0.8085	0.0833	0.0021	0.0093	0.0616	0.5510
	9	-0.0059	-0.0000	-0.0009	-0.0009	0.0215	0.8085	0.0836	0.0010	0.0092		
104	7	0.600	-0.001	-0.000	-0.000	0.1121	2.00	-2.00	-2.00	2.01	0.0741	0.8025
	8	0.0115	0.0002	0.0015	0.0015	0.0408	0.6552	-0.0049	-0.0000	0.0007	0.1171	0.6291
	9	-0.0075	-0.0000	-0.0011	-0.0011	0.0408	0.7362	0.0123	0.0002	0.0015		
105	7	0.599	-3.17	-0.000	-0.000	0.1489	4.99	-5.07	-5.04	4.99	0.1700	0.6545
	8	0.0243	0.0007	0.0033	0.0033	0.1123	0.6796	-0.0482	-0.0016	0.0063	-0.1888	0.5367
	9	-0.0215	-0.0004	-0.0030	-0.0030	0.1123	0.7039	0.0026	-0.0000	0.0002		
105	7	0.600	-1.07	-0.000	-0.000	0.1457	4.99	-5.03	-5.04	5.01	0.1496	0.6835
	8	0.0258	0.0007	0.0034	0.0034	0.1234	0.6678	-0.0273	-0.0008	0.0037	0.1210	0.6533
	9	-0.0214	-0.0005	-0.0030	-0.0030	0.1234	0.7064	0.0191	0.0004	0.0025		
105	7	0.600	-0.005	-0.000	-0.000	0.1475	5.00	-5.02	-5.04	5.01	0.1407	0.6912
	8	0.0272	0.0008	0.0036	0.0036	0.1215	0.6695	-0.0201	-0.0005	0.0027	0.1263	0.6484
	9	-0.0207	-0.0005	-0.0029	-0.0029	0.1215	0.7054	0.0248	0.0006	0.0032		
105	7	0.600	0.46	-0.000	-0.000	0.1468	4.99	-5.01	-5.04	5.03	0.1328	0.6918
	8	0.0263	0.0007	0.0035	0.0035	0.1164	0.6744	-0.0146	-0.0003	0.0020	0.1390	0.6616
	9	-0.0210	-0.0004	-0.0029	-0.0029	0.1164	0.6974	0.0312	0.0008	0.0041		
105	7	0.600	1.00	-0.000	-0.000	0.1444	4.99	-5.00	-5.04	5.03	0.1234	0.7316
	8	0.0258	0.0007	0.0034	0.0034	0.1262	0.6756	-0.0105	-0.0002	0.0015	0.1356	0.6534
	9	-0.0215	-0.0005	-0.0030	-0.0030	0.1262	0.7049	0.0365	0.0009	0.0047		
105	7	0.601	3.08	-0.000	-0.000	0.1440	5.00	-4.98	-5.03	5.06	0.5562	0.4745
	8	0.0256	0.0007	0.0034	0.0034	0.1212	0.6690	0.0025	0.0002	0.0002	0.1503	0.6301
	9	-0.0199	-0.0004	-0.0028	-0.0028	0.1212	0.7113	0.0525	0.0015	0.0066		
105	7	0.600	6.17	-0.000	-0.000	0.1344	5.00	-4.92	-5.04	5.09	0.2140	0.6347
	8	0.0250	0.0006	0.0033	0.0033	0.1241	0.6630	0.0298	0.0012	0.0037	0.1470	0.5783
	9	-0.0196	-0.0004	-0.0028	-0.0028	0.1241	0.7179	0.0732	0.0021	0.0084		

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key														
105	8	0.600	9.28	0.0006	0.0033	0.1272	0.1333	4.99	0.6592	0.7385	-4.88	0.0550	-5.04	5.09	0.2025	0.6148
	6	0.0254	0.0006	-0.0004	-0.0027	0.0033	0.1333	0.6592	0.7385	0.0850	0.0550	0.0022	0.0094	0.0067	0.0868	0.5556
	9	-0.0184	-0.0004	-0.0004	-0.0027	0.0033	0.1333	0.7385	0.7385	0.0850	0.0550	0.0014	0.0094	0.0067	0.0868	0.5556
105	9	0.600	12.42	0.0006	0.0034	0.1269	0.1383	4.99	0.6542	0.7367	-4.85	0.0758	5.06	0.1775	0.5766	
	6	0.0260	0.0006	-0.0005	-0.0028	0.0034	0.1383	0.6542	0.7367	0.0811	0.0758	0.0026	0.0089	0.0087	0.0397	0.5493
	9	-0.0196	-0.0005	-0.0005	-0.0028	0.0034	0.1383	0.7367	0.7367	0.0811	0.0758	0.0006	0.0089	0.0087	0.0397	0.5493
105	10	0.600	-0.07	0.0007	-0.0034	0.1442	0.1210	5.00	0.6723	0.7045	-5.01	-0.0192	5.02	0.1461	0.6969	
	6	0.0258	0.0007	-0.0005	-0.0034	0.1442	0.1210	0.6723	0.7045	-0.0272	-0.0192	-0.0005	-0.0026	0.0026	0.1303	0.6578
	9	-0.0214	-0.0005	-0.0005	-0.0034	0.1210	0.1210	0.7045	0.7045	0.0272	0.0192	0.0007	0.0026	0.0026	0.1303	0.6578
106	11	0.802	-0.01	0.0000	-0.0005	-0.0333	2.5437	0.04	0.6898	1.3838	3.03	0.0199	0.02	3.02	0.2103	0.6747
	6	0.0038	-0.0000	0.0000	0.0005	-0.0333	2.5437	0.04	0.6898	1.3838	0.0199	0.0199	0.0008	0.0026	0.1060	0.6374
	9	0.0006	0.0003	0.0003	0.0001	-0.0333	2.5437	1.3838	1.3838	0.0197	0.0197	0.0004	0.0026	0.0026	0.1060	0.6374
106	12	0.799	-0.01	0.0000	-0.0005	-0.0276	1.5213	0.04	0.7238	0.7269	3.03	0.0195	3.02	0.2101	0.6731	
	6	0.0037	-0.0000	0.0000	0.0005	-0.0276	1.5213	0.04	0.7238	0.7269	0.0195	0.0195	0.0008	0.0026	0.1076	0.6323
	9	0.0010	0.0003	0.0003	0.0001	-0.0276	1.5213	0.7269	0.7269	0.0193	0.0193	0.0004	0.0026	0.0026	0.1076	0.6323
107	1	0.798	0.00	0.0000	-0.0005	-0.0252	1.8119	0.04	0.6909	0.9898	5.05	0.0316	5.04	0.1785	0.6696	
	6	0.0039	-0.0000	0.0000	0.0005	-0.0252	1.8119	0.04	0.6909	0.9898	0.0316	0.0316	0.0011	0.0042	0.1120	0.6658
	9	0.0009	0.0003	0.0003	0.0001	-0.0252	1.8119	0.9898	0.9898	0.0308	0.0308	0.0006	0.0041	0.0041	0.1120	0.6658
107	2	0.797	-0.00	0.0000	-0.0005	-0.0209	1.5532	0.04	0.7015	0.8609	5.04	0.0315	5.04	0.1778	0.6701	
	6	0.0039	-0.0000	0.0000	0.0005	-0.0209	1.5532	0.04	0.7015	0.8609	0.0315	0.0315	0.0011	0.0042	0.1085	0.6597
	9	0.0010	0.0003	0.0003	0.0001	-0.0209	1.5532	0.8609	0.8609	0.0307	0.0307	0.0006	0.0040	0.0040	0.1085	0.6597
108	1	0.800	-0.03	0.0008	-0.0040	0.1347	0.1016	5.01	0.6761	0.6846	-5.03	-0.0207	5.04	0.1424	0.7131	
	6	0.0298	0.0008	-0.0004	0.0030	0.1347	0.1016	0.6761	0.6846	0.0279	-0.0207	-0.0005	-0.0029	0.0029	0.1139	0.6585
	9	-0.0223	-0.0004	-0.0004	-0.0030	0.1016	0.1016	0.6846	0.6846	0.0279	0.0279	0.0006	0.0036	0.0036	0.1139	0.6585
108	2	0.801	-0.03	0.0008	-0.0040	0.1341	0.1025	5.01	0.6770	0.6966	-5.03	-0.0212	5.03	0.1445	0.7161	
	6	0.0301	0.0008	-0.0004	0.0030	0.1341	0.1025	0.6770	0.6966	0.0278	-0.0212	-0.0006	-0.0036	0.0036	0.1168	0.6640
	9	-0.0219	-0.0004	-0.0004	-0.0030	0.1025	0.1025	0.6966	0.6966	0.0278	0.0278	0.0006	0.0036	0.0036	0.1168	0.6640
279	3	0.601	0.00	0.0000	0.0004	-0.0557	0.6241	-0.02	0.6963	0.0153	2.98	0.0153	3.00	0.2425	0.7105	
	6	0.0033	-0.0000	0.0000	0.0004	-0.0557	0.6241	-0.02	0.6963	0.0153	2.98	0.0153	0.0021	0.0225	0.6712	
	9	0.0014	0.0001	-0.0000	0.0000	0.6241	-0.1041	-0.1041	-0.1041	0.0174	0.0174	0.0004	0.0023	0.0023	0.6712	
279	4	0.601	0.00	0.0000	0.0004	-0.0460	0.6070	-0.02	0.7002	0.0157	2.98	0.0157	3.07	0.2384	0.7077	
	6	0.0032	-0.0000	0.0000	0.0004	-0.0460	0.6070	-0.02	0.7002	0.0157	2.98	0.0157	0.0022	0.0225	0.6757	
	9	0.0013	0.0001	-0.0000	0.0000	0.6070	-0.2295	-0.2295	-0.2295	0.0176	0.0176	0.0004	0.0023	0.0023	0.6757	

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key															
279	5	0.601	-0.0004	0.0004	-0.0553	-0.7082	-0.03	-2.95	0.06	-2.98	0.1082	0.6645					
	6	0.0032	-0.0000	0.0004	-0.0769	0.7082	-0.0134	-0.0134	-0.0002	-0.0017	0.2336	0.6880					
	9	0.0012	-0.0001	-0.0000	-0.0553	-0.1615	-0.0134	-0.0134	-0.0006	-0.0018							
279	6	0.601	-0.0000	0.0004	-0.0598	-0.7047	-0.02	-2.94	0.06	-2.98	0.1157	0.6771					
	6	0.0033	-0.0000	0.0004	-0.08130	0.7047	-0.0134	-0.0134	-0.0003	-0.0018	0.2353	0.6951					
	9	0.0012	-0.0002	-0.0000	-0.08130	-0.1336	-0.0134	-0.0134	-0.0006	-0.0018							
281	2	0.600	-0.0001	0.0000	-0.0315	-0.02	1.01	0.06	0.06	0.96	0.3971	0.7257					
	6	0.0035	-0.0000	0.0005	-1.2943	0.7188	0.0048	0.0055	0.0003	0.0007	0.0438	0.6837					
	9	0.0009	-0.0002	0.0000	-1.2943	0.4174	0.0055	0.0000	0.0000	0.0007							
281	3	0.799	-0.0002	0.0005	-0.0275	-0.01	-0.95	0.06	0.06	-1.00	0.0193	0.7401					
	6	0.0035	-0.0000	0.0005	-2.1373	0.7082	-0.0040	-0.0047	-0.0003	-0.0006	0.3947	0.7297					
	9	0.0005	-0.0002	0.0000	-2.1373	0.3911	-0.0047	-0.0047	-0.0003	-0.0006							
281	4	0.796	-0.0000	0.0004	-0.0555	-0.01	5.05	0.06	0.06	5.04	0.1700	0.6738					
	6	0.0035	-0.0000	0.0004	-1.6350	0.6801	0.288	0.009	0.009	0.0038	0.1127	0.6767					
	9	0.0009	-0.0003	0.0000	-1.6350	0.4970	0.294	0.006	0.006	0.0039							
281	5	0.799	-0.0005	0.0004	-0.0418	-0.01	-5.07	0.06	0.06	-5.06	0.1440	0.6969					
	6	0.0034	-0.0000	0.0004	-4.2391	0.6843	-0.0254	-0.007	-0.007	-0.0035	0.2008	0.6975					
	9	0.0004	-0.0003	0.0001	-4.2391	1.3203	-0.0261	-0.0010	-0.0010	-0.0036							
281	6	0.600	-0.0001	0.0007	0.0888	1.01	0.06	1.02	0.06	0.03	2.1073	-0.0822					
	6	0.0083	-0.0001	0.0011	0.5240	0.6885	0.0003	0.0001	0.0001	0.0000	-2.4488	0.1429					
	9	0.0051	-0.0005	0.0007	0.5240	0.6894	0.0003	-0.0001	-0.0001	0.0000							
281	7	0.799	-0.0002	0.0005	3.8916	-1.03	-0.06	-1.02	0.06	0.03	0.9552	0.4848					
	6	0.0002	-0.0001	0.0000	-0.2205	1.3136	0.0008	0.0001	0.0001	0.0000	-0.9597	0.3853					
	9	-0.0033	-0.0001	-0.0005	-0.2205	0.7989	0.0007	-0.0001	-0.0001	0.0000							
281	8	0.600	-0.0008	0.0044	0.1243	5.07	0.06	5.05	0.06	0.04	0.6682	0.3951					
	6	0.0330	-0.0006	0.0044	0.2056	0.6678	0.0011	0.0001	0.0001	0.0000	-0.4139	0.4996					
	9	0.0277	-0.0011	0.0037	0.2056	0.6839	0.0013	-0.0001	-0.0001	0.0001							
281	9	0.600	-0.0009	0.0030	0.2189	-5.03	-0.06	-5.05	0.06	0.03	0.6257	0.2542					
	6	0.0220	-0.0004	0.0034	0.0982	0.6972	0.0010	0.0001	0.0001	0.0000	-1.7384	0.1462					
	9	-0.0248	-0.0004	-0.0034	0.0982	0.6959	0.0005	-0.0001	-0.0001	0.0000							
281	10	0.799	-0.0004	0.0023	0.1221	3.04	-2.96	-3.01	0.06	2.95	0.1663	0.7209					
	6	0.0183	-0.0004	0.0023	0.0607	0.6502	-0.0116	-0.0003	0.0003	0.0016	0.1193	0.6545					
	9	-0.0128	-0.0001	-0.0017	0.0607	0.6727	-0.0147	-0.0003	-0.0003	0.0019							

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																								
201	11	7	0.799	-0.002	-0.001	0.1330	0.6771	-5.07	-0.0224	-5.05	-0.0007	5.03	0.1557	0.7063	8	0.0299	0.0008	0.0040	0.0946	0.6837	0.0256	-0.0006	-0.0034	0.1180	0.6692	
		9	-0.0222	-0.0004	-0.0030																					
202	3	7	0.799	0.004	44.99	0.1250	0.6891	-0.009	0.0001	-3.00	0.0001	-0.0001	-0.5353	0.8210	8	0.0171	0.0004	0.0023	0.0832	0.6517	0.0002	-0.0001	-0.0000	-3.6500	-1.1259	
		9	-0.0120	-0.0002	-0.0015																					
202	4	7	0.799	0.01	45.00	0.2524	0.6356	-0.06	0.0000	2.96	0.0000	-0.0003	-0.1750	0.7990	8	0.0123	0.0006	0.0015	0.2745	0.7020	0.0013	-0.0001	0.0000	-0.5460	0.3596	
		9	-0.0150	-0.0000	-0.0021																					
202	5	7	0.799	0.01	45.00	-1.2165	0.4229	-2.95	-0.0120	0.06	0.0003	-0.0017	0.1490	0.7203	8	0.0007	0.0001	0.0000	3.6140	0.3288	0.0178	0.0003	-0.0023	0.1111	0.6591	
		9	0.0003	0.0002	0.0000																					
203	6	7	1.249	-3.11	0.00	-0.7390	0.072	-0.10	-0.255	0.06	0.0000	-0.0034	0.0060	0.6679	8	0.0011	0.0001	0.0002	1.2356	0.2386	-0.0259	-0.0001	-0.0035	0.0201	0.6803	
		9	0.0007	0.0001	0.0000																					
203	7	7	1.249	-0.00	0.00	-0.6140	0.7642	-0.07	-0.002	0.07	0.0001	-0.0000	2.3313	0.2910	8	0.0015	0.0001	0.0002	1.4956	0.1697	-0.0002	-0.0002	-0.0000	6.6997	1.0091	
		9	0.0000	0.0002	0.0000																					
203	8	7	1.251	1.04	0.00	-0.4795	0.730	-0.05	-0.002	0.06	0.0000	0.0011	-0.0264	0.6774	8	0.0017	0.0001	0.0002	1.5141	0.730	0.0002	-0.0002	0.0010	-0.1053	0.6612	
		9	0.0000	0.0002	0.0000																					
203	9	7	1.249	3.14	0.00	-0.6607	0.7313	-0.04	-0.254	0.06	0.0001	-0.0033	-0.0197	0.6639	8	0.0011	0.0001	0.0000	1.2790	0.4404	0.0230	-0.0003	0.0031	-0.0705	0.6641	
		9	0.0010	0.0002	0.0000																					
203	10	7	1.251	6.26	0.00	-1.7694	0.9404	-0.01	-0.502	0.06	0.0003	-0.0066	-0.0300	0.6621	8	0.0006	0.0002	0.0001	1.0748	0.6643	0.0516	-0.0005	0.0068	-0.0574	0.6657	
		9	0.0014	0.0003	0.0001																					
203	11	7	1.252	9.30	0.00	-3.6204	1.8691	0.00	0.753	0.06	0.0006	0.0098	-0.0431	0.6527	8	0.0002	0.0001	0.0001	1.5047	0.8774	0.0767	-0.0008	0.0099	-0.0568	0.6507	
		9	0.0011	0.0003	0.0001																					
00 203	12	7	1.250	12.61	0.00	-1.0129	0.9211	0.03	0.973	0.06	0.0008	0.0123	-0.0430	0.6364	8	0.0004	0.0001	0.0001	3.4377	1.1606	0.0978	-0.0011	0.0123	-0.0610	0.6330	
		9	0.0004	0.0003	0.0001																					

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key											
283	15	7	1.250	-0.001	0.002	-0.4265	-0.02	-0.06	0.06	-0.05	12.4622	4.7748	
		6	0.0021	-0.0001	0.0002	-2.2742	0.6894	-0.0000	-0.0001	-0.0000	13.2065	1.2447	
		9	0.0007	0.0003	0.0001	0.0000	0.7350	-0.0004	-0.0003	-0.0001			
284	1	7	1.251	0.02	-0.00	-0.5980	-0.02	6.03	0.06	6.02	-0.0357	0.6821	
		8	0.0014	-0.0001	0.0002	-3.1781	0.7686	0.0302	-0.0002	0.0041	-0.0722	0.6900	
		9	0.0005	0.0003	0.0000	0.0000	0.7843	0.0309	-0.0004	0.0042			
284	2	7	1.248	-3.07	-0.00	-0.9322	-0.02	6.00	0.06	5.99	-0.1437	0.7806	
		8	0.0009	-0.0001	0.0001	-5.2353	0.7234	0.0049	-0.0001	0.0007	-0.2828	0.7519	
		9	0.0002	0.0002	0.0000	0.0000	0.3781	0.0052	-0.0002	0.0007			
284	3	7	1.250	0.03	-0.00	-0.6628	-0.02	6.03	0.06	6.01	-0.0370	0.6811	
		8	0.0014	-0.0001	0.0002	-1.9914	0.7349	0.0305	-0.0002	0.0041	-0.0738	0.6912	
		9	0.0008	0.0003	0.0001	0.0000	0.6397	0.0313	-0.0004	0.0043			
284	4	7	1.249	1.07	0.00	-0.5018	-0.02	6.04	0.06	6.02	-0.0397	0.6768	
		8	0.0016	-0.0001	0.0002	-1.9916	0.6979	0.0401	-0.0003	0.0054	-0.0731	0.6822	
		9	0.0008	0.0003	0.0000	0.0000	0.5485	0.0398	-0.0005	0.0054			
284	5	7	1.248	3.18	0.00	-0.4866	-0.02	6.05	0.06	6.03	-0.0479	0.6679	
		8	0.0015	-0.0001	0.0002	-2.2480	0.6811	0.0583	-0.0007	0.0077	-0.0666	0.6756	
		9	0.0007	0.0003	0.0000	0.0000	0.6614	0.0568	-0.0007	0.0076			
284	6	7	1.248	6.30	0.00	-1.8895	-0.02	6.08	0.06	6.05	-0.0479	0.6540	
		8	0.0025	-0.0002	0.0001	-1.5796	1.0140	0.0815	-0.0007	0.0106	-0.0617	0.6519	
		9	0.0012	0.0003	0.0002	0.0000	0.8185	0.0822	-0.0010	0.0107			
284	7	7	1.249	9.44	0.00	-2.5956	-0.02	6.10	0.06	6.06	-0.0457	0.6372	
		8	0.0004	-0.0002	0.0000	-1.6057	1.2201	0.1027	-0.0009	0.0130	-0.0598	0.6302	
		9	0.0010	0.0003	0.0002	0.0000	0.9893	0.1033	-0.0012	0.0130			
284	8	7	1.249	12.65	0.00	-1.0080	-0.02	6.12	0.06	6.06	-0.0395	0.6181	
		8	0.0009	-0.0001	0.0001	-4.3233	0.9378	0.1187	-0.0009	0.0146	-0.0665	0.6158	
		9	0.0004	0.0003	0.0001	0.0000	1.3404	0.1198	-0.0015	0.0147			
284	9	7	1.249	0.03	0.00	-0.5642	-0.02	6.02	0.06	6.02	-0.0423	0.6701	
		8	0.0018	-0.0002	0.0002	-4.2463	0.6261	0.0306	-0.0002	0.0041	-0.0599	0.6843	
		9	0.0004	0.0003	0.0000	0.0000	0.8410	0.0312	-0.0003	0.0042			
285	1	7	1.250	-3.09	45.00	-0.0525	-0.05	6.01	0.04	6.01	-0.0511	0.6901	
		8	-0.0185	-0.0001	-0.0024	0.0491	0.6650	0.0128	-0.0001	0.0017	-0.1137	0.6819	
		9	-0.0159	0.0001	-0.0021	-0.0491	0.6742	0.0125	-0.0002	0.0017			

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key →											
285	2	1.249 0.0003 0.0005	0.06 -0.0001 0.0003	44.99 0.0001 0.0000	-2.9650 2.9408	1.6925 0.8387	-0.03 0.0301	6.04 0.0311	0.06 -0.0002	0.06 0.0042	-0.0437 0.0465	0.6732 0.6824	
285	3	1.250 0.0057 0.0060	1.07 -0.0001 0.0003	44.99 0.0008 0.0008	-0.1268 0.3309	0.7202 0.7068	-0.01 0.07	6.04 0.0359	0.07 -0.0003	6.02 0.0048	-0.0485 -0.0557	0.6713 0.6783	
285	4	1.250 0.0189 0.0190	3.16 -0.0001 0.0003	44.99 0.0025 0.0026	-0.0343 0.0885	-0.00 0.6863	0.09 0.0474	6.04 0.0496	0.09 -0.0005	6.02 0.0067	-0.0556 -0.0620	0.6692 0.6752	
285	5	1.249 0.0368 0.0396	6.27 -0.0002 0.0001	44.99 0.0050 0.0054	-0.0423 0.0155	0.6836 0.6861	0.11 -0.0008	6.06 0.0682	0.11 -0.0008	6.04 0.0090	-0.0601 -0.0607	0.6608 0.6622	
285	6	1.248 0.0534 0.0577	9.42 -0.0005 -0.0001	44.99 0.0072 0.0078	-0.0495 -0.0096	0.6823 0.6769	0.13 -0.0009	6.07 0.0802	0.13 -0.0009	6.05 0.0105	-0.0597 -0.0575	0.6548 0.6468	
285	7	1.247 0.0689 0.0758	12.59 -0.0007 -0.0003	44.99 0.0092 0.0100	-0.0544 -0.0229	0.6739 0.6611	0.14 -0.0011	6.08 0.1022	0.14 -0.0011	6.06 0.0120	-0.0607 -0.0542	0.6513 0.6330	
285	8	1.247 0.0004 0.0005	0.05 -0.0001 0.0003	44.99 0.0001 0.0000	-2.1012 3.4590	-0.04 1.2953	0.06 -0.0003	6.03 0.0314	0.06 -0.0003	6.02 0.0043	-0.0479 -0.0512	0.6608 0.6826	
287	3	1.250 0.0011 0.0006	-3.00 -0.0001 0.0002	0.00 0.0001 0.0000	-0.7109 1.6423	-0.04 0.7863	0.06 -0.0005	15.05 0.0501	0.06 -0.0005	15.02 0.0069	-0.0533 -0.0797	0.6970 0.6827	
287	4	1.250 0.0022 0.0010	0.06 -0.0001 0.0002	0.00 0.0003 0.0001	-0.3728 1.3124	-0.04 0.7247	0.06 -0.0007	15.07 0.0760	0.06 -0.0007	15.03 0.0102	-0.0506 -0.0671	0.6708 0.6614	
287	5	1.250 0.0017 0.0007	1.12 -0.0001 0.0002	0.00 0.0002 0.0000	-0.4652 1.9886	-0.03 0.7138	0.06 -0.0011	15.08 0.0843	0.06 -0.0008	15.04 0.0112	-0.0491 -0.0688	0.6642 0.6528	
287	6	1.250 0.0011 0.0008	3.24 -0.0001 0.0002	0.00 0.0001 0.0000	-0.5529 1.7161	-0.04 0.7868	0.06 -0.0013	15.10 0.0987	0.06 -0.0009	15.05 0.0126	-0.0475 -0.0660	0.6525 0.6403	

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																			
267	7	1.249	6.36	0.00	-2.4708	-0.03	15.12	0.06	15.07	-0.0463	0.6347	0.0003	0.0001	0.0001	1.4653	0.1169	-0.0010	0.0148	0.0143	-0.0535	0.6153
267	8	1.248	9.46	0.00	-9.3473	-0.03	15.14	0.06	15.07	-0.0349	0.6142	0.0001	0.0001	0.0001	4.0798	0.1281	-0.0008	0.0157	0.0159	-0.0624	0.6053
267	9	1.250	12.65	0.00	-1.0684	-0.03	15.14	0.05	15.07	-0.0365	0.5984	0.0003	0.0000	0.0000	1.0312	0.1320	-0.0009	0.0158	0.0161	-0.0601	0.5940
267	10	1.251	0.07	0.00	-0.4477	-0.03	15.07	0.06	15.04	-0.0531	0.6657	0.0019	0.0001	0.0002	0.7153	0.0759	-0.0008	0.0101	0.0102	-0.0678	0.6592
268	1	1.250	-3.03	45.00	0.0568	-0.05	15.06	0.03	15.03	-0.0472	0.6849	-0.0164	-0.0024	0.6570	0.0592	0.0596	-0.0005	0.0081	0.0080	-0.0803	0.6744
268	2	1.249	0.05	45.00	-3.7624	-0.03	15.07	0.06	15.03	-0.0538	0.6670	-0.0004	0.0000	-4.9043	0.0749	-0.0008	0.0099	0.0107	0.0109	-0.0651	0.6589
268	3	1.248	1.12	45.00	0.1090	-0.02	15.07	0.07	15.04	-0.0547	0.6623	0.0054	0.0008	0.7436	0.0810	-0.0008	0.0107	0.0109	-0.0668	0.6525	
268	4	1.249	5.18	44.99	-0.0305	-0.00	15.09	0.08	15.05	-0.0561	0.6586	0.0190	0.0025	0.6694	0.0900	-0.0010	0.0118	0.0121	-0.0623	0.6438	
268	5	1.250	6.33	44.99	-0.0404	0.01	15.09	0.10	15.06	-0.0593	0.6482	0.0370	0.0050	0.6812	0.1035	-0.0012	0.0134	0.0136	-0.0567	0.6286	
268	6	1.249	9.48	44.99	-0.0539	0.03	15.10	0.12	15.07	-0.0473	0.6376	0.0549	0.0074	0.6819	0.1141	-0.0013	0.0145	0.0147	-0.0473	0.6121	
268	7	1.251	12.65	45.00	-0.0524	0.05	15.11	0.14	15.07	-0.0560	0.6392	0.0704	0.0095	0.6761	0.1206	-0.0013	0.0154	0.0156	-0.0473	0.6020	
		0.0797	-0.0005	0.0105	-0.0370	0.6591	0.1301	-0.0012	0.0156	-0.0473	0.6020	0.0797	0.0105	0.6591	0.1301	-0.0012	0.0156	0.0156	-0.0473	0.6020	

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																		
288	8	9	1.250	-0.0001	0.0002	0.0003	0.0001	0.0003	45.000	0.0000	0.0000	4.7546	-1.1870	-0.02	15.07	0.0764	-0.0008	0.06	0.0101	-0.0567	0.6617
			0.0002	0.0003	0.0001	0.0003	0.0001	0.0003	0.0000	0.0000	5.2872	1.0769	0.0781	-0.0010	0.0102	0.0781	-0.0010	0.0102	-0.0657	0.6556	
289	1	6	1.250	-0.0187	-0.0158	0.0001	-0.0002	-0.0001	45.000	-0.0025	-0.0021	0.0548	0.6670	-0.05	-0.09	-0.0169	-0.0002	-0.05	0.0688	0.6994	
			0.0158	0.0001	0.0001	0.0001	0.0001	0.0001	0.0025	0.0021	-0.0408	0.6772	-0.0183	-0.0002	-0.0025	-0.0169	-0.0002	-0.0025	0.0583	0.7060	
289	2	7	1.249	-0.0000	0.0004	0.0002	-0.0001	0.0002	45.000	0.0000	0.0000	10.7289	-3.9204	-0.03	-0.08	-0.0003	-0.0002	0.06	3.8772	2.5978	
			0.0004	0.0002	0.0002	0.0001	0.0002	0.0000	0.0000	0.0000	3.4812	0.6095	-0.0000	-0.0002	-0.0001	-0.0003	-0.0002	0.06	722.0829	0.6617	
289	3	7	1.249	0.0064	0.0065	0.0003	-0.0001	0.0003	45.000	0.0009	0.0009	-0.1002	0.7017	-0.02	-0.07	0.0041	-0.0002	0.07	-0.2419	0.5977	
			0.0065	0.0003	0.0003	0.0003	0.0001	0.0003	0.0009	0.0009	0.2778	0.6951	0.0050	-0.0002	0.0005	0.0041	-0.0002	0.07	-0.3136	0.5758	
289	4	7	1.250	0.0251	0.0263	0.0002	-0.0001	0.0002	45.000	0.0034	0.0036	-0.0285	0.6791	0.00	-0.04	0.0206	-0.0001	0.09	-0.0479	0.6465	
			0.0263	0.0002	0.0002	0.0002	0.0001	0.0002	0.0034	0.0036	0.0400	0.6847	0.0232	-0.0001	0.0029	0.0206	-0.0001	0.09	-0.0870	0.6397	
289	5	7	1.250	0.0366	0.0395	0.0000	-0.0002	0.0000	45.000	0.0050	0.0054	-0.0376	0.6841	0.01	-0.03	0.0329	-0.0002	0.10	-0.0449	0.6528	
			0.0395	0.0000	0.0000	0.0000	0.0002	0.0000	0.0050	0.0054	0.0110	0.6841	0.0350	0.01	0.0329	-0.0002	0.10	0.0046	-0.0652	0.6607	
289	6	7	1.249	0.0537	0.0578	0.0001	-0.0001	0.0001	44.999	0.0073	0.0078	-0.0471	0.6826	0.03	-0.01	0.0500	-0.0004	0.12	-0.0475	0.6582	
			0.0578	0.0001	0.0001	0.0001	0.0001	0.0001	0.0073	0.0078	-0.0134	0.6753	0.0552	0.03	0.0500	-0.0004	0.12	0.0073	-0.0582	0.6620	
289	7	7	1.249	0.0681	0.0751	0.0003	-0.0003	0.0003	44.999	0.0092	0.0099	-0.0522	0.6759	0.04	-0.00	0.0653	-0.0006	0.14	-0.0504	0.6643	
			0.0751	0.0003	0.0003	0.0003	0.0003	0.0003	0.0092	0.0099	-0.0206	0.6629	0.0740	0.04	0.0653	-0.0006	0.14	0.0096	-0.0530	0.6526	
289	8	7	1.248	0.0001	0.0000	0.0003	-0.0001	0.0003	45.000	0.0000	0.0000	5.5849	-1.1267	-0.03	-0.07	-0.0005	-0.0002	0.06	2.3033	1.6131	
			0.0000	0.0003	0.0003	0.0003	0.0003	0.0000	0.0000	0.0000	-60.7278	-0.0002	-0.0002	-0.0002	-0.0001	-0.0002	-0.0002	0.06	7.0738	4.0554	
290	1	7	0.899	-0.0196	-0.0186	0.0001	-0.0001	0.0001	45.000	0.0025	-0.0025	0.1915	0.6487	-0.05	-0.10	-0.0211	-0.0004	0.03	0.1151	0.6932	
			0.0196	0.0001	0.0001	0.0001	0.0001	0.0001	0.0025	-0.0025	0.0311	0.6878	-0.0216	-0.0004	-0.0030	-0.0004	-0.0006	0.03	0.1606	0.7006	
290	2	7	0.901	0.0013	0.0005	0.0004	-0.0001	0.0004	44.999	0.0003	0.0001	-0.4719	1.1848	-0.02	-0.08	-0.0014	0.0000	0.06	-0.1638	0.9513	
			0.0005	0.0004	0.0004	0.0004	0.0004	0.0004	0.0003	0.0001	-3.8094	0.9593	-0.0010	-0.0010	-0.0002	-0.0014	0.0000	0.06	-1.4326	0.4222	

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd																		
290	5	7	0.856	1.03	44.99	0.1270	-0.02	-0.06	0.07	-0.03	0.3846	0.5609							
		6	0.0066	0.0001	0.0010	0.4376	0.7890	0.0034	0.0002	0.0003	0.0128	0.5894							
		9	0.0072	0.0006	0.0009		0.6835	0.0060	0.0000	0.0007									
290	4	7	0.899	3.10	44.99	0.0617	0.00	-0.05	0.09	-0.01	0.1966	0.7089							
		6	0.0240	0.0002	0.0033	0.1772	0.6913	0.0146	0.0005	0.0020	0.0604	0.6889							
		9	0.0237	0.0008	0.0032		0.6767	0.0205	0.0002	0.0028									
290	5	7	0.900	6.19	44.99	0.0710	0.02	-0.01	0.11	0.00	0.0986	0.6807							
		6	0.0454	0.0006	0.0061	0.1398	0.6746	0.0364	0.0007	0.0049	0.0769	0.6821							
		9	0.0449	0.0012	0.0057		0.6439	0.0422	0.0006	0.0057									
290	6	7	0.897	9.29	44.99	0.0934	0.04	0.00	0.14	0.01	0.0916	0.6436							
		6	0.0582	0.0010	0.0074	0.1163	0.6382	0.0524	0.0009	0.0067	0.0760	0.6177							
		9	0.0617	0.0014	0.0075		0.6145	0.0593	0.0009	0.0073									
290	7	7	0.896	12.40	44.99	0.0859	0.06	0.01	0.15	0.02	0.0888	0.6403							
		6	0.0721	0.0012	0.0085	0.0977	0.6180	0.0669	0.0011	0.0085	0.0626	0.5902							
		9	0.0762	0.0014	0.0089		0.5885	0.0753	0.0009	0.0089									
290	8	7	0.898	-0.05	44.99	-0.6289	-0.03	-0.07	0.06	-0.04	-0.0794	0.9691							
		6	0.0012	-0.0001	0.0002	7.6750	1.0909	-0.0015	0.0000	-0.0003	-1.2444	0.4080							
		9	0.0002	0.0004	0.0000		0.8067	0.0011	-0.0002	0.0000									
291	1	7	0.899	-3.07	-0.00	-0.2248	-0.02	-0.10	0.06	-0.07	0.0848	0.6732							
		6	0.0027	-0.0001	0.0003	-7.2059	0.6911	-0.0301	-0.0005	-0.0040	0.1134	0.6813							
		9	-0.0002	0.0003	-0.0000		1.8069	-0.0316	-0.0007	-0.0043									
291	2	7	0.901	-0.02	-0.00	-0.0604	-0.02	-0.06	0.06	-0.04	7.3015	1.0758							
		6	0.0027	-0.0000	0.0004	48.5720	0.7572	0.0001	0.0001	0.0001	-1.5631	0.7131							
		9	0.0000	0.0003	-0.0000		-12.1258	0.0009	-0.0002	0.0001									
291	3	7	0.901	1.02	-0.00	-0.0429	-0.02	-0.05	0.06	-0.03	0.3752	0.7295							
		6	0.0030	-0.0000	0.0004	9.1045	0.6525	0.0075	0.0005	0.0010	0.0722	0.6697							
		9	0.0002	0.0003	-0.0000		-1.2547	0.0081	0.0001	0.0010									
291	4	7	0.899	5.11	0.00	0.0074	-0.02	-0.02	0.06	-0.01	0.1074	0.6916							
		6	0.0025	0.0000	0.0004	4.7924	0.7865	0.0310	0.0006	0.0042	0.0497	0.6768							
		9	0.0003	0.0003	-0.0000		-0.4588	0.0312	0.0003	0.0042									
291	5	7	0.898	6.18	-0.00	-0.2269	-0.02	0.00	0.05	0.01	0.0850	0.6253							
		6	0.0018	-0.0000	0.0003	3.2788	0.8364	0.0563	0.0009	0.0070	0.0728	0.6253							
		9	0.0005	0.0003	0.0000		0.4599	0.0577	0.0008	0.0072									

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																				
291	6	7	0.0017	0.0006	0.0001	0.0002	-0.3979	2.8315	-0.02	0.7414	0.7997	0.03	0.783	0.0791	0.06	0.0011	0.0008	0.02	0.0093	0.0758	0.0542	0.5944	0.5841
291	7	7	0.0018	0.0002	0.0001	0.0002	-0.3025	6.7732	-0.02	0.782	1.8513	0.03	0.912	0.0904	0.06	0.0104	0.0104	0.03	0.0104	0.0616	0.0187	0.5707	0.5763
291	8	7	0.0028	0.0002	0.0003	-0.0004	-0.0586	6.7974	-0.02	0.7847	-0.4437	-0.06	0.005	0.0011	0.06	-0.0001	0.0001	0.04	0.0001	1.5490	-1.1257	0.3817	0.5672
292	1	7	0.0027	-0.0002	0.0001	0.0003	-0.2248	-6.3609	-0.02	0.6911	1.8583	0.01	0.063	0.0076	0.06	0.0004	0.0010	5.99	0.0009	0.3408	0.0387	0.7642	0.6804
292	2	7	0.0024	-0.0003	0.0003	-0.0003	-0.0688	4.2287	-0.02	0.8111	-0.1208	0.04	0.350	0.0400	0.06	0.0006	0.0055	6.02	0.0049	0.0977	0.0366	0.7061	0.6883
292	3	7	0.0022	-0.0003	0.0003	-0.0003	-0.1761	4.6919	-0.02	0.7552	-0.4596	0.05	0.433	0.0494	0.06	0.0008	0.0065	6.04	0.0058	0.0993	0.0581	0.6728	0.6666
292	4	7	0.0021	-0.0005	0.0003	-0.0003	-0.0562	2.7445	-0.02	0.7767	0.1889	0.07	0.611	0.0626	0.06	0.0010	0.0078	6.05	0.0078	0.0871	0.0747	0.6393	0.6279
292	5	7	0.0020	-0.0006	0.0003	0.0002	-0.1538	2.0175	-0.02	0.7119	0.4393	0.09	0.814	0.0811	0.06	0.0013	0.0094	6.06	0.0095	0.0816	0.0523	0.5884	0.5828
292	6	7	0.0018	-0.0010	0.0003	0.0003	-0.2568	1.8435	-0.02	0.8885	0.8958	0.10	0.921	0.0897	0.06	0.0011	0.0102	6.06	0.0105	0.0618	0.0158	0.5697	0.5720
292	7	7	0.0020	-0.0006	0.0003	0.0001	-0.2758	2.7091	-0.02	0.6526	0.8436	0.09	0.981	0.0964	0.06	0.0004	0.0112	6.05	0.0112	-0.0227	-0.0068	0.5739	0.5750
292	8	7	0.0025	-0.0005	0.0003	-0.0003	-0.0835	2.8442	-0.02	0.7568	0.0304	0.04	0.348	0.0396	0.06	0.0006	0.0054	6.01	0.0048	0.0969	0.0435	0.7004	0.6870

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																			
293	1	7	0.901	-5.04	44.99	0.1634	-0.05	6.02	0.03	6.00	0.2193	0.7476	0.0202	-0.0006	-0.0026	0.0131	0.0005	0.0019	0.0114	0.0024	
			-0.0180	-0.0002	-0.0024	0.0556	0.6828	0.0174	0.0000	0.0019	0.0114	0.6988									
293	2	7	0.899	0.01	44.99	-0.8729	-0.03	6.03	0.06	6.02	0.0994	0.7081	0.0008	-0.0001	0.0002	0.0327	0.0006	0.0046	0.0443	0.0055	
			0.0011	0.0003	0.0001	1.5216	1.4073	0.0401	0.0003	0.0006	0.0443	0.6857									
293	3	7	0.900	1.07	44.99	0.1451	-0.02	6.04	0.07	6.03	0.0975	0.6916	0.0055	0.0001	0.0009	0.0464	0.0005	0.0052	0.0571	0.0062	
			0.0055	0.0001	0.0009	0.4288	0.7090	0.0464	0.0005	0.0052	0.0571	0.6685									
293	4	7	0.900	5.13	44.99	0.0070	-0.00	6.05	0.09	6.04	0.0873	0.6639	0.0254	0.0000	0.0035	0.0484	0.0008	0.0071	0.0737	0.0071	
			0.0232	0.0008	0.0032	0.1908	0.6895	0.0555	0.0008	0.0071	0.0737	0.6409									
293	5	7	0.899	6.21	44.99	0.0463	0.02	6.07	0.12	6.06	0.0663	0.6381	0.0463	0.0012	0.0062	0.0666	0.0008	0.0084	0.0617	0.0084	
			0.0443	0.0012	0.0057	0.1402	0.6475	0.0697	0.0008	0.0084	0.0617	0.6033									
293	6	7	0.896	9.32	45.00	0.0843	0.04	6.09	0.14	6.06	0.0700	0.6154	0.0590	0.0009	0.0075	0.0805	0.0010	0.0093	0.0635	0.0093	
			0.0625	0.0015	0.0076	0.1209	0.6144	0.0804	0.0010	0.0093	0.0635	0.5783									
293	7	7	0.902	12.42	45.00	0.0654	0.06	6.09	0.15	6.06	0.0504	0.6013	0.0747	0.0009	0.0092	0.0936	0.0009	0.0112	0.0304	0.0112	
			0.0778	0.0014	0.0092	0.0900	0.5919	0.0873	0.0009	0.0112	0.0304	0.5769									
293	8	7	0.898	0.04	44.99	-0.5746	-0.02	6.03	0.06	6.03	0.0962	0.7014	0.0009	-0.0001	0.0002	0.0335	0.0006	0.0046	0.0494	0.0046	
			0.0007	0.0003	0.0001	2.4969	1.0942	0.0403	0.0006	0.0046	0.0494	0.6830									
294	1	7	0.900	-2.98	-0.00	-0.2748	-0.02	15.06	0.06	15.04	0.0754	0.6810	0.0019	-0.0001	0.0002	0.0545	0.0008	0.0074	0.0520	0.0074	
			0.0000	0.0002	-0.0000	12.4987	-3.9289	0.0624	0.0008	0.0074	0.0520	0.6264									
294	2	7	0.901	0.04	0.00	-0.1685	-0.02	15.08	0.06	15.05	0.0545	0.6439	0.0023	-0.0000	0.0003	0.0797	0.0008	0.0102	0.0347	0.0102	
			0.0001	0.0002	-0.0000	13.1889	3.0377	0.0789	0.0008	0.0102	0.0347	0.5857									
294	3	7	0.900	1.09	0.00	-0.0920	-0.02	15.09	0.06	15.06	0.0606	0.6170	0.0024	-0.0002	0.0000	0.0857	0.0010	0.0098	0.0263	0.0098	
			0.0001	0.0002	-0.0000	7.9321	-1.3605	0.0837	0.0010	0.0098	0.0263	0.5851									

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key															
294	4	0.899	0.0030	0.0004	0.0000	0.00	-0.0586	-0.7397	0.0911	0.06	15.09	0.0006	0.0106	0.0335	0.0241	0.5955	0.5731
	7	0.0004	0.0000	0.0000	0.0000	0.00	3.5427	0.0503	0.0888	0.0004	0.0911	0.0006	0.0106	0.0241	0.0241	0.5955	0.5731
294	5	0.899	0.0026	0.0003	0.0001	0.00	-0.1795	-0.7460	0.0969	0.06	15.10	0.0007	0.0110	0.0394	0.0022	0.5719	0.5707
	7	0.0009	0.0003	0.0001	0.0001	0.00	1.9528	0.6293	0.0954	0.0000	0.0969	0.0000	0.0109	0.0022	0.0022	0.5719	0.5707
294	6	0.900	0.0020	0.0001	0.0001	0.00	-0.2745	-0.7217	0.1029	0.06	15.10	0.0005	0.0117	0.0261	0.0107	0.5709	0.5697
	7	0.0010	0.0003	0.0001	0.0001	0.00	1.5304	0.7558	0.1002	-0.0002	0.1029	0.0002	0.0114	0.0261	0.0107	0.5709	0.5697
294	7	0.901	0.0019	0.0001	0.0001	0.00	-0.3601	-0.7089	0.1065	0.06	15.10	0.0000	0.0122	0.0039	0.0178	0.5759	0.5653
	7	0.0007	0.0001	0.0001	0.0001	0.00	2.0779	0.9048	0.1057	-0.0003	0.1065	0.0000	0.0119	0.0039	0.0178	0.5759	0.5653
294	8	0.898	0.0023	0.0000	0.0000	0.00	-0.0944	-0.8338	0.0789	0.06	15.09	0.0009	0.0096	0.0594	0.0357	0.5138	0.5846
	7	0.0001	0.0000	0.0000	0.0000	0.00	8.6073	1.9531	0.0786	0.0005	0.0789	0.0005	0.0091	0.0594	0.0357	0.5138	0.5846
295	1	0.899	0.0681	0.0004	0.0001	4.99	0.0363	14.78	15.07	15.07	15.07	15.07	15.05	0.0507	0.0494	0.6267	0.6235
	7	0.0681	0.0004	0.0001	0.0001	4.99	0.0788	0.6266	0.0683	15.07	0.0687	0.0006	0.0085	0.0507	0.0494	0.6267	0.6235
295	2	0.898	0.0788	0.0010	0.0009	4.99	0.0365	14.79	15.09	15.08	15.09	15.08	15.06	0.0399	0.0455	0.6234	0.5913
	7	0.0788	0.0010	0.0009	0.0009	4.99	0.0662	0.5999	0.0793	15.08	0.0825	0.0007	0.0093	0.0399	0.0455	0.6234	0.5913
295	3	0.898	0.0874	0.0006	0.0006	4.99	0.0377	14.80	15.09	15.09	15.09	15.09	15.06	0.0399	0.0382	0.6158	0.5896
	7	0.0874	0.0006	0.0006	0.0006	4.99	0.0762	0.5900	0.0842	15.09	0.0868	0.0006	0.0099	0.0399	0.0382	0.6158	0.5896
295	4	0.899	0.0957	0.0007	0.0102	4.99	0.0377	14.81	15.10	15.09	15.10	15.09	15.07	0.0377	0.0374	0.6025	0.5817
	7	0.0957	0.0007	0.0102	0.0102	4.99	0.0586	0.5860	0.0895	15.09	0.0933	0.0006	0.0112	0.0377	0.0374	0.6025	0.5817
295	5	0.900	0.1039	0.0003	0.0108	4.99	0.0159	14.81	15.10	15.08	15.10	15.08	15.06	0.0148	0.0139	0.5940	0.5753
	7	0.1039	0.0003	0.0108	0.0108	4.99	0.0296	0.5822	0.0967	15.08	0.1034	0.0003	0.0122	0.0148	0.0139	0.5940	0.5753
295	6	0.899	0.1036	0.0000	0.0116	45.00	0.0011	14.80	15.10	15.09	15.10	15.09	15.06	0.0262	0.0071	0.5810	0.5707
	7	0.1036	0.0000	0.0116	0.0116	45.00	0.0185	0.5831	0.1009	15.09	0.1072	0.0005	0.0124	0.0262	0.0071	0.5810	0.5707

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd																			
295	7	0.901	12.47	45.00	-0.0116	14.80	15.10	15.09	15.07	0.0029	0.5764									
	8	0.1078	-0.0002	0.0125	0.0178	0.5798	0.1126	0.0000	0.0129	0.0022	0.5701									
	9	0.1035	0.0003	0.0120	0.0178	0.5804	0.1061	0.0000	0.0121	0.0022	0.5701									
295	8	0.900	0.11	45.00	0.0329	14.79	15.09	15.08	15.06	0.0347	0.6213									
	8	0.0840	0.0005	0.0103	0.0685	0.6143	0.0838	0.0005	0.0104	0.0418	0.5891									
	9	0.0792	0.0010	0.0096	0.0685	0.6059	0.0800	0.0006	0.0094	0.0418	0.5891									
296	1	0.898	-3.03	44.99	0.0521	6.03	6.01	6.07	6.01	0.2351	0.7373									
	8	0.0160	0.0001	0.0023	0.2276	0.7272	0.0138	0.0006	0.0020	0.0295	0.6871									
	9	0.0146	0.0006	0.0020	0.2276	0.7070	0.0165	0.0000	0.0022	0.0295	0.6871									
296	2	0.899	0.06	44.99	0.0155	6.06	6.03	6.09	6.02	0.0717	0.6945									
	8	0.0414	0.0001	0.0058	0.1167	0.7089	0.0362	0.0005	0.0050	0.0579	0.6852									
	9	0.0377	0.0008	0.0051	0.1167	0.6786	0.0391	0.0004	0.0053	0.0579	0.6852									
296	3	0.899	1.09	44.99	0.0514	6.06	6.04	6.10	6.03	0.0883	0.6582									
	8	0.0473	0.0004	0.0065	0.1271	0.6879	0.0396	0.0007	0.0052	0.0631	0.6642									
	9	0.0433	0.0011	0.0056	0.1271	0.6522	0.0454	0.0005	0.0060	0.0631	0.6642									
296	4	0.899	3.16	44.99	0.0671	6.08	6.04	6.12	6.05	0.0688	0.6431									
	8	0.0575	0.0007	0.0073	0.1225	0.6418	0.0510	0.0007	0.0065	0.0771	0.6441									
	9	0.0556	0.0013	0.0070	0.1225	0.6289	0.0553	0.0008	0.0071	0.0771	0.6441									
296	5	0.899	6.24	44.99	0.0555	6.10	6.06	6.13	6.06	0.0565	0.6248									
	8	0.0740	0.0008	0.0091	0.1012	0.6164	0.0684	0.0007	0.0085	0.0688	0.6248									
	9	0.0723	0.0014	0.0087	0.1012	0.6037	0.0704	0.0009	0.0085	0.0688	0.6248									
296	6	0.901	9.33	44.99	0.0490	6.11	6.08	6.14	6.07	0.0589	0.6089									
	8	0.0864	0.0008	0.0104	0.0829	0.6023	0.0826	0.0009	0.0100	0.0652	0.5823									
	9	0.0811	0.0013	0.0093	0.0829	0.5784	0.0823	0.0010	0.0095	0.0652	0.5823									
296	7	0.899	12.44	44.99	0.0331	6.11	6.09	6.14	6.06	0.0489	0.5891									
	8	0.0936	0.0006	0.0109	0.0430	0.5861	0.0924	0.0009	0.0108	0.0489	0.5891									
	9	0.0864	0.0007	0.0100	0.0430	0.5816	0.0881	0.0004	0.0101	0.0489	0.5891									
296	8	0.898	0.06	44.99	0.0186	6.05	6.03	6.09	6.03	0.0829	0.6316									
	8	0.0416	0.0001	0.0058	0.1202	0.7005	0.0357	0.0005	0.0048	0.0487	0.6762									
	9	0.0380	0.0009	0.0051	0.1202	0.6756	0.0393	0.0003	0.0053	0.0487	0.6762									
297	1	1.248	-3.04	45.00	-0.0839	6.02	6.00	6.06	5.99	-0.0590	0.6899									
	8	0.0119	-0.0002	0.0017	0.0788	0.7268	0.0122	-0.0001	0.0016	-0.1272	0.6899									
	9	0.0121	0.0001	0.0017	0.0788	0.7041	0.0120	-0.0003	0.0016	-0.1272	0.6899									

← See Key →

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key										
297	2	7	1.251	0.0003	44.99	-0.0485	6.05	6.02	6.08	6.01	-0.0413	0.6740
		6	0.0312	-0.0000	0.0044	0.0042	0.7079	0.0290	-0.0002	0.0039	-0.0598	0.6852
		9	0.0313	0.0000	0.0043	0.0043	0.6947	0.0313	-0.0003	0.0042		
297	3	7	1.248	1.11	44.99	-0.0458	6.05	6.02	6.08	6.01	-0.0443	0.6690
		8	0.0379	-0.0003	0.0053	-0.0030	0.7060	0.0351	-0.0003	0.0047	-0.0613	0.6810
		9	0.0380	-0.0000	0.0053	-0.0030	0.6965	0.0381	-0.0004	0.0052		
297	4	7	1.249	3.20	44.99	-0.0505	6.07	6.03	6.10	6.02	-0.0531	0.6746
		8	0.0507	-0.0005	0.0069	-0.0152	0.6881	0.0458	-0.0004	0.0066	-0.0648	0.6787
		9	0.0517	-0.0001	0.0071	-0.0152	0.6875	0.0501	-0.0006	0.0068		
297	5	7	1.250	6.33	44.99	-0.0578	6.08	6.04	6.12	6.03	-0.0553	0.6647
		8	0.0669	-0.0007	0.0090	-0.0218	0.6764	0.0639	-0.0007	0.0085	-0.0554	0.6679
		9	0.0712	-0.0003	0.0094	-0.0218	0.6659	0.0686	-0.0007	0.0091		
297	6	7	1.250	6.33	44.99	-0.0581	6.08	6.05	6.12	6.03	-0.0543	0.6670
		8	0.0674	-0.0007	0.0091	-0.0199	0.6778	0.0634	-0.0006	0.0084	-0.0547	0.6701
		9	0.0714	-0.0002	0.0095	-0.0199	0.6692	0.0681	-0.0007	0.0091		
297	7	7	1.249	6.29	44.99	-0.0580	6.08	6.05	6.12	6.03	-0.0550	0.6666
		8	0.0673	-0.0007	0.0091	-0.0201	0.6760	0.0638	-0.0007	0.0085	-0.0554	0.6665
		9	0.0713	-0.0002	0.0095	-0.0201	0.6674	0.0688	-0.0007	0.0091		
297	8	7	1.248	9.48	44.99	-0.0603	6.10	6.06	6.14	6.04	-0.0571	0.6593
		8	0.0818	-0.0009	0.0109	-0.0242	0.6665	0.0787	-0.0009	0.0103	-0.0534	0.6516
		9	0.0676	-0.0004	0.0114	-0.0242	0.6517	0.0867	-0.0009	0.0113		
297	9	7	1.251	12.64	44.99	-0.0584	6.11	6.07	6.15	6.06	-0.0574	0.6573
		8	0.0927	-0.0010	0.0121	-0.0263	0.6528	0.0915	-0.0010	0.0120	-0.0550	0.6524
		9	0.1023	-0.0005	0.0129	-0.0263	0.6336	0.1034	-0.0011	0.0130		
297	10	7	1.250	0.08	44.99	-0.0495	6.04	6.02	6.08	6.01	-0.0463	0.6685
		8	0.0312	-0.0003	0.0044	0.0104	0.7050	0.0290	-0.0002	0.0038	-0.0607	0.6862
		9	0.0312	0.0000	0.0043	0.0104	0.7006	0.0309	-0.0003	0.0042		
298	1	7	1.251	-2.94	44.99	-0.0554	15.08	15.06	15.06	15.02	-0.0490	0.6793
		8	0.0601	-0.0006	0.0083	-0.0400	0.6929	0.0599	-0.0005	0.0081	-0.0762	0.6783
		9	0.0584	-0.0004	0.0080	-0.0400	0.6922	0.0597	-0.0009	0.0081		
298	2	7	1.251	0.18	44.99	-0.0573	15.10	15.07	15.08	15.04	-0.0512	0.6665
		8	0.0774	-0.0008	0.0104	-0.0349	0.6737	0.0758	-0.0007	0.0101	-0.0632	0.6559
		9	0.0773	-0.0005	0.0104	-0.0349	0.6727	0.0786	-0.0009	0.0103		

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																				
298	3	7	1.250	1.222	4.99	15.11	15.08	15.09	15.04	15.07	-0.0521	0.6658	0.0831	-0.0009	0.0110	0.0667	0.0804	-0.0008	0.0107	-0.0621	0.6654	
		8	0.0632	-0.0005	0.0110	0.6640	0.0841	-0.0010	0.0110	0.0110	0.0621	0.6554										
29A	4	7	1.250	5.28	4.99	15.12	15.08	15.11	15.04	15.04	-0.0559	0.6579	0.0930	-0.0010	0.0121	0.6530	0.0945	-0.0011	0.0121	-0.0627	0.6438	
		8	0.0942	-0.0005	0.0122	0.6505	0.0945	-0.0011	0.0122	0.0121	0.0627	0.6438										
29A	5	7	1.250	6.42	4.99	15.13	15.09	15.13	15.05	15.05	-0.0610	0.6474	0.1052	-0.0012	0.0134	0.6343	0.1040	-0.0012	0.0134	-0.0582	0.6250	
		8	0.1101	-0.0006	0.0136	0.6287	0.1106	-0.0012	0.0136	0.0136	0.0582	0.6250										
298	6	7	1.250	9.55	4.99	15.13	15.10	15.14	15.06	15.06	-0.0605	0.6382	0.1147	-0.0015	0.0144	0.6279	0.1145	-0.0013	0.0144	-0.0560	0.6104	
		8	0.1234	-0.0008	0.0152	0.6188	0.1244	-0.0013	0.0152	0.0151	0.0560	0.6104										
298	7	7	1.250	12.70	4.99	15.14	15.11	15.15	15.08	15.08	-0.0577	0.6358	0.1223	-0.0016	0.0153	0.6268	0.1216	-0.0014	0.0153	-0.0398	0.5946	
		8	0.1300	-0.0005	0.0157	0.6038	0.1293	-0.0010	0.0153	0.0153	0.0398	0.5946										
298	8	7	1.249	0.17	4.99	15.10	15.07	15.08	15.03	15.03	-0.0543	0.6614	0.0774	-0.0009	0.0104	0.6718	0.0762	-0.0008	0.0103	-0.0636	0.6574	
		8	0.0778	-0.0005	0.0104	0.6710	0.0786	-0.0010	0.0103	0.0103	0.0636	0.6574										
299	3	7	0.599	-3.09	-0.00	-0.02	-0.10	0.06	-0.06	-0.06	0.1548	0.6466	0.0021	-0.0000	0.0003	0.7882	-0.0250	-0.0007	-0.0032	0.2184	0.6474	
		8	0.0003	0.0001	0.0000	0.2497	-0.0250	-0.0010	0.0003	0.0003	0.1548	0.6466										
299	4	7	0.599	-0.04	-0.00	-0.02	-0.06	0.05	-0.03	-0.03	-2.0480	0.3307	0.0023	-0.0000	0.0003	0.7603	-0.0003	0.0001	-0.0001	-0.5003	0.7161	
		8	0.0001	0.0001	0.0000	0.3357	-0.0010	-0.0001	0.0001	0.0001	0.3307	0.7161										
299	5	7	0.600	1.01	-0.00	-0.03	-0.05	0.05	-0.03	-0.03	0.2898	0.7257	0.0025	-0.0001	0.0003	0.7678	-0.0074	0.0004	0.0009	0.0722	0.6642	
		8	-0.0001	0.0001	-0.0000	0.9480	0.0074	0.0001	0.0001	0.0001	0.2898	0.6642										
299	6	7	0.600	5.10	0.00	-0.02	-0.02	0.06	-0.00	-0.00	0.2032	0.6672	0.0024	-0.0000	0.0010	0.8002	-0.0257	0.0010	0.0034	0.1362	0.6538	
		8	0.0001	0.0001	0.0000	1.5348	0.0246	0.0006	0.0006	0.0032	0.1362	0.6538										
299	7	7	0.600	6.19	-0.00	-0.02	0.02	0.06	0.03	0.03	0.1851	0.6348	0.0013	-0.0000	0.0016	0.8001	0.0508	0.0016	0.0064	0.1590	0.6355	
		8	0.0004	0.0002	0.0001	1.4662	0.0528	0.0016	0.0016	0.0067	0.1590	0.6355										

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt. Cd See Key

299	8	7 8 9	0.599 0.0019 0.0010	7.28 -0.0000 0.0002	0.00 0.0003 0.0002	-0.1939 1.2165	-0.02 0.8091 1.1409	0.05 0.0720 0.0735	0.06 0.0023 0.0022	0.06 0.0085 0.0085	0.1600 0.1510	0.5937 0.5817
299	9	7 8 9	0.598 0.0014 -0.0002	12.43 -0.0001 0.0001	0.00 0.0001 0.0000	-0.3977 -3.0922	-0.02 0.6831 -0.5469	0.05 0.0824 0.0853	0.05 0.0019 0.0016	0.05 0.0092 0.0095	0.1195 0.0978	0.5612 0.5591
299	10	7 8 9	0.598 0.0029 0.0000	-0.003 -0.0000 0.0002	-0.00 0.0003 0.0000	-0.0551 15.6105	-0.02 0.6467 5.7991	-0.06 -0.0003 -0.0004	0.06 0.0001 -0.0001	-0.04 -0.0000 -0.0000	-1.9236 -1.8611	0.5779 0.5822
300	1	7 8 9	0.599 -0.0154 -0.0146	-3.12 -0.0007 -0.0003	45.00 -0.0018 -0.0018	0.2445 0.1265	-0.05 0.6149 0.6300	-0.09 -0.0161 -0.0152	0.03 -0.0004 -0.0007	-0.05 -0.0021 -0.0020	0.1531 0.2525	0.6553 0.6696
300	2	7 8 9	0.599 0.0016 0.0008	-0.02 -0.0000 0.0002	44.99 0.0003 0.0001	-0.1783 1.6387	-0.02 1.0046 1.1988	-0.06 -0.0028 0.0004	0.06 0.0000 -0.0001	-0.0003 -0.0000 -0.0000	-0.0270 -1.5095	0.6985 0.6388
300	3	7 8 9	0.598 0.0068 0.0050	1.00 0.0001 0.0004	44.99 0.0009 0.0007	0.0992 0.4277	-0.01 0.7804 0.7800	-0.06 0.0035 0.0059	0.06 0.0002 0.0000	-0.002 0.0005 0.0007	0.3995 0.0572	0.7046 0.6377
300	4	7 8 9	0.599 0.0193 0.0185	3.07 0.0006 0.0009	44.99 0.0025 0.0025	0.1600 0.2492	0.00 0.6875 0.6834	-0.04 0.0132 0.0161	0.09 0.0006 0.0003	-0.01 0.0018 0.0020	0.2354 0.1233	0.6809 0.6404
300	5	7 8 9	0.599 0.0381 0.0397	6.16 0.0012 0.0015	44.99 0.0050 0.0052	0.1620 0.1949	0.03 0.6625 0.6556	-0.01 0.0311 0.0344	0.12 0.0011 0.0009	0.01 0.0041 0.0044	0.1843 0.1408	0.6665 0.6524
300	6	7 8 9	0.599 0.0541 0.0564	9.26 0.0019 0.0021	44.99 0.0069 0.0070	0.1794 0.1906	0.06 0.6369 0.6259	0.01 0.0494 0.0547	0.15 0.0017 0.0017	0.04 0.0063 0.0068	0.1816 0.1615	0.6460 0.6270
300	7	7 8 9	0.599 0.0688 0.0704	12.38 0.0023 0.0025	44.99 0.0083 0.0082	0.1698 0.1813	0.08 0.6067 0.5892	0.03 0.0639 0.0692	0.18 0.0021 0.0021	0.06 0.0078 0.0081	0.1661 0.1552	0.6177 0.5900
300	8	7 8 9	0.598 0.0020 0.0003	-0.00 -0.0000 0.0002	44.99 0.0003 0.0001	-0.1744 -3.6308	-0.03 0.8192 2.1777	-0.06 -0.0006 -0.0010	0.06 0.0001 -0.0001	-0.03 -0.0000 -0.0000	-0.8365 -0.6025	0.7017 0.2923

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key													
301	1	7	0.600	-5.03	-0.00	-0.02	6.00	0.05	6.00	0.3148	0.0276	0.600	0.8285		
		6	0.0018	-0.0000	0.0002	0.7774	0.0053	0.0003	0.0008						
		9	-0.0003	0.0001	-0.0000	0.3893	0.0069	0.0000	0.0009				0.6900		
301	2	7	0.600	0.01	-0.00	-0.02	6.04	0.05	6.03	0.1844	0.1285	0.604	0.6806		
		6	0.0029	-0.0000	0.0003	0.6705	0.0328	0.0012	0.0044						
		9	-0.0000	0.0001	0.0000	-1.7306	0.0338	0.0008	0.0045				0.6733		
301	3	7	0.598	1.03	-0.00	-0.02	6.06	0.05	6.04	0.1791	0.1371	0.605	0.6609		
		6	0.0025	-0.0000	0.0004	0.7847	0.0424	0.0015	0.0056						
		9	-0.0003	0.0001	-0.0000	0.5092	0.0436	0.0011	0.0058				0.6678		
301	4	7	0.600	3.13	-0.00	-0.02	6.09	0.05	6.07	0.1750	0.1509	0.607	0.6283		
		6	0.0021	-0.0000	0.0003	0.7557	0.0581	0.0020	0.0073						
		9	-0.0009	0.0001	-0.0000	0.5115	0.0590	0.0017	0.0073				0.6213		
301	5	7	0.600	6.22	0.00	-0.02	6.10	0.06	6.09	0.1505	0.1396	0.609	0.5936		
		6	0.0021	-0.0000	0.0003	0.7359	0.0763	0.0022	0.0090						
		9	0.0009	0.0002	0.0001	1.3419	0.0761	0.0021	0.0088				0.5786		
301	6	7	0.599	9.28	-0.00	-0.02	6.10	0.06	6.08	0.1048	0.0814	0.608	0.5621		
		6	0.0017	-0.0000	0.0002	0.7884	0.0834	0.0017	0.0093						
		9	0.0005	0.0002	0.0001	1.4966	0.0867	0.0014	0.0096				0.5586		
301	7	7	0.600	12.44	0.00	-0.02	6.09	0.05	6.07	0.0649	0.0401	0.607	0.5596		
		6	0.0015	-0.0001	0.0002	0.7168	0.0832	0.0010	0.0093						
		9	0.0001	0.0001	0.0001	4.4155	0.0812	0.0006	0.0090				0.5570		
301	8	7	0.600	0.00	-0.00	-0.02	6.04	0.05	6.04	0.1818	0.1251	0.604	0.6703		
		6	0.0031	-0.0000	0.0004	0.6703	0.0327	0.0011	0.0043						
		9	-0.0002	0.0001	0.0000	-0.5469	0.0330	0.0008	0.0044				0.6736		
302	9	7	0.599	-3.06	44.99	6.02	6.01	6.05	6.01	0.2240	0.0970	6.01	0.7162		
		6	0.0134	0.0003	0.0019	0.7119	0.0138	0.0006	0.0019						
		9	0.0112	0.0005	0.0016	0.7344	0.0130	0.0002	0.0020				0.6694		
302	10	7	0.600	0.04	44.99	6.06	6.04	6.09	6.03	0.1793	0.1271	6.03	0.6741		
		6	0.0343	0.0010	0.0046	0.6814	0.0295	0.0010	0.0039						
		9	0.0325	0.0012	0.0044	0.6845	0.0317	0.0008	0.0042				0.6708		
302	11	7	0.599	1.07	44.99	6.05	6.04	6.10	6.04	0.1749	0.1366	6.04	0.6715		
		6	0.0406	0.0012	0.0054	0.6679	0.0353	0.0012	0.0047						
		9	0.0389	0.0014	0.0052	0.6680	0.0391	0.0010	0.0052				0.6742		

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																						
302	12	7	0.599	5.14	44.99	0.0017	0.0068	0.1667	6.08	6.06	6.12	6.06	0.1710	0.6557	0.0528	0.0017	0.0065	0.1937	0.6426	0.476	0.0016	0.0062	0.1463	0.6449	
		6	0.528	0.0019	0.0065	0.0019	0.0065	0.1937	0.6426	0.476	0.0016	0.0062	0.1463	0.6449	0.0528	0.0017	0.0065	0.1937	0.6426	0.476	0.0016	0.0062	0.1463	0.6449	0.0528
		9	0.0507	0.0019	0.0065	0.0019	0.0065	0.1937	0.6426	0.476	0.0016	0.0062	0.1463	0.6449	0.0507	0.0019	0.0065	0.1937	0.6426	0.476	0.0016	0.0062	0.1463	0.6449	0.0507
302	13	7	0.600	6.23	44.99	0.0021	0.0084	0.1552	6.11	6.08	6.15	6.08	0.1581	0.6279	0.0688	0.0024	0.0082	0.1836	0.6081	0.620	0.0019	0.0077	0.1510	0.6046	0.0688
		8	0.688	0.0021	0.0084	0.0021	0.0084	0.1552	0.6081	0.620	0.0019	0.0077	0.1510	0.6046	0.0688	0.0024	0.0082	0.1836	0.6081	0.620	0.0019	0.0077	0.1510	0.6046	0.0688
		9	0.0680	0.0024	0.0082	0.0024	0.0082	0.1836	0.6081	0.620	0.0019	0.0077	0.1510	0.6046	0.0680	0.0024	0.0082	0.1836	0.6081	0.620	0.0019	0.0077	0.1510	0.6046	0.0680
302	14	7	0.599	9.31	44.99	0.0022	0.0087	0.1445	6.12	6.10	6.15	6.09	0.1470	0.6057	0.0759	0.0022	0.0087	0.1489	0.5779	0.769	0.0022	0.0088	0.1365	0.5765	0.0759
		8	0.775	0.0022	0.0087	0.0022	0.0087	0.1445	0.5779	0.769	0.0022	0.0088	0.1365	0.5765	0.0759	0.0022	0.0087	0.1489	0.5779	0.769	0.0022	0.0088	0.1365	0.5765	0.0759
		9	0.0759	0.0022	0.0087	0.0022	0.0087	0.1489	0.5779	0.769	0.0022	0.0088	0.1365	0.5765	0.0759	0.0022	0.0087	0.1489	0.5779	0.769	0.0022	0.0088	0.1365	0.5765	0.0759
302	15	7	0.598	12.42	44.99	0.0019	0.0092	0.1116	6.13	6.12	6.15	6.08	0.1465	0.5911	0.0857	0.0016	0.0092	0.1025	0.5729	0.876	0.0016	0.0094	0.1265	0.5624	0.0857
		8	0.857	0.0016	0.0092	0.0016	0.0092	0.1116	0.5729	0.876	0.0016	0.0094	0.1265	0.5624	0.0857	0.0016	0.0092	0.1025	0.5729	0.876	0.0016	0.0094	0.1265	0.5624	0.0857
		9	0.0810	0.0016	0.0092	0.0016	0.0092	0.1025	0.5729	0.876	0.0016	0.0094	0.1265	0.5624	0.0810	0.0016	0.0092	0.1025	0.5729	0.876	0.0016	0.0094	0.1265	0.5624	0.0810
302	16	7	0.599	0.03	44.99	0.0010	0.0043	0.1536	6.05	6.04	6.09	6.03	0.1624	0.6769	0.0335	0.0012	0.0043	0.1892	0.6798	0.305	0.0011	0.0044	0.1265	0.6669	0.0335
		8	0.335	0.0012	0.0043	0.0010	0.0043	0.1536	0.6798	0.305	0.0011	0.0044	0.1265	0.6669	0.0335	0.0012	0.0043	0.1892	0.6798	0.305	0.0011	0.0044	0.1265	0.6669	0.0335
		9	0.0321	0.0012	0.0043	0.0012	0.0043	0.1892	0.6798	0.305	0.0011	0.0044	0.1265	0.6669	0.0321	0.0012	0.0043	0.1892	0.6798	0.305	0.0011	0.0044	0.1265	0.6669	0.0321
303	1	7	0.600	-2.97	44.99	0.0019	0.0076	0.1571	15.12	15.09	15.10	15.07	0.1614	0.6425	0.0616	0.0018	0.0076	0.1538	0.6375	0.619	0.0016	0.0076	0.1312	0.6164	0.0616
		8	0.616	0.0018	0.0076	0.0019	0.0076	0.1571	0.6375	0.619	0.0016	0.0076	0.1312	0.6164	0.0616	0.0018	0.0076	0.1538	0.6375	0.619	0.0016	0.0076	0.1312	0.6164	0.0616
		9	0.0602	0.0018	0.0076	0.0018	0.0076	0.1538	0.6375	0.619	0.0016	0.0076	0.1312	0.6164	0.0602	0.0018	0.0076	0.1538	0.6375	0.619	0.0016	0.0076	0.1312	0.6164	0.0602
303	2	7	0.599	0.12	44.99	0.0021	0.0089	0.1445	15.14	15.11	15.12	15.09	0.1397	0.6142	0.0758	0.0024	0.0089	0.1656	0.6037	0.752	0.0020	0.0088	0.1344	0.5884	0.0758
		8	0.758	0.0021	0.0089	0.0021	0.0089	0.1445	0.6037	0.752	0.0020	0.0088	0.1344	0.5884	0.0758	0.0021	0.0089	0.1656	0.6037	0.752	0.0020	0.0088	0.1344	0.5884	0.0758
		9	0.0737	0.0024	0.0089	0.0024	0.0089	0.1656	0.6037	0.752	0.0020	0.0088	0.1344	0.5884	0.0737	0.0024	0.0089	0.1656	0.6037	0.752	0.0020	0.0088	0.1344	0.5884	0.0737
303	3	7	0.600	1.12	44.99	0.0021	0.0088	0.1363	15.14	15.11	15.12	15.09	0.1370	0.6094	0.0752	0.0023	0.0088	0.1576	0.5883	0.779	0.0019	0.0089	0.1267	0.5753	0.0752
		8	0.778	0.0021	0.0088	0.0021	0.0088	0.1363	0.5883	0.779	0.0019	0.0089	0.1267	0.5753	0.0752	0.0023	0.0088	0.1576	0.5883	0.779	0.0019	0.0089	0.1267	0.5753	0.0752
		9	0.0752	0.0023	0.0088	0.0023	0.0088	0.1576	0.5883	0.779	0.0019	0.0089	0.1267	0.5753	0.0752	0.0023	0.0088	0.1576	0.5883	0.779	0.0019	0.0089	0.1267	0.5753	0.0752
303	4	7	0.599	3.19	44.99	0.0018	0.0091	0.1087	15.14	15.12	15.12	15.10	0.1359	0.5962	0.0829	0.0018	0.0091	0.1249	0.5684	0.846	0.0017	0.0093	0.1060	0.5618	0.0829
		8	0.829	0.0018	0.0091	0.0018	0.0091	0.1087	0.5684	0.846	0.0017	0.0093	0.1060	0.5618	0.0829	0.0018	0.0091	0.1249	0.5684	0.846	0.0017	0.0093	0.1060	0.5618	0.0829
		9	0.0799	0.0019	0.0091	0.0019	0.0091	0.1249	0.5684	0.846	0.0017	0.0093	0.1060	0.5618	0.0799	0.0019	0.0091	0.1249	0.5684	0.846	0.0017	0.0093	0.1060	0.5618	0.0799
303	5	7	0.599	6.24	44.99	0.0012	0.0094	0.0693	15.14	15.11	15.11	15.08	0.1065	0.5695	0.0871	0.0014	0.0094	0.0860	0.5723	0.878	0.0012	0.0098	0.1065	0.5604	0.0871
		8	0.881	0.0012	0.0094	0.0012	0.0094	0.0693	0.5723	0.878	0.0012	0.0098	0.1065	0.5604	0.0881	0.0014	0.0094	0.0860	0.5723	0.878	0.0012	0.0098	0.1065	0.5604	0.0881
		9	0.0826	0.0014	0.0094	0.0014	0.0094	0.0860	0.5723	0.878	0.0012	0.0098	0.1065	0.5604	0.0826	0.0014	0.0094	0.0860	0.5723	0.878	0.0012	0.0098	0.1065	0.5604	0.0826
303	6	7	0.599	9.32	44.99	0.0006	0.0089	0.0516	15.13	15.10	15.10	15.08	0.0576	0.5660	0.0816	0.0011	0.0089	0.0723	0.5689	0.816	0.0009	0.0090	0.0576	0.5660	0.0816
		8	0.852	0.0006	0.0089	0.0006	0.0089	0.0516	0.5689	0.816	0.0009	0.0090	0.0576	0.5660	0.852	0.0006	0.0089	0.0723	0.5689	0.816	0.0009	0.0090	0.0576	0.5660	0.852
		9	0.0785	0.0011	0.0089	0.0011	0.0089	0.0723	0.5689	0.816	0.0009	0.0090	0.0576	0.5660	0.0785	0.0011	0.0089	0.0723	0.5689	0.816	0.0009	0.0090	0.0576	0.5660	0.0785

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key														
303	7	0.599	12.41	45.00	0.0325	15.12	15.09	15.07	0.0416	0.0097	0.0005	0.0860	0.0826	0.0092	0.0416	0.5646
	6	0.0847	0.0005	0.0094	0.0478	0.5589	0.0860	0.0007	0.0005	0.0007	0.0005	0.0860	0.0826	0.0092	0.0416	0.5646
	9	0.0820	0.0007	0.0094	0.0478	0.5589	0.0860	0.0007	0.0005	0.0007	0.0005	0.0860	0.0826	0.0092	0.0416	0.5646
303	8	0.600	0.08	44.99	0.1467	15.14	15.11	15.10	0.1407	0.0088	0.0020	0.0722	0.0754	0.0088	0.1407	0.6120
	6	0.0752	0.0022	0.0089	0.1639	0.6004	0.0722	0.0020	0.0020	0.0088	0.0020	0.0722	0.0754	0.0088	0.1407	0.6120
	9	0.0757	0.0024	0.0088	0.1639	0.6004	0.0722	0.0020	0.0020	0.0088	0.0020	0.0722	0.0754	0.0088	0.1407	0.6120
304	1	0.600	-3.02	-0.00	-0.1939	-0.02	15.08	15.06	0.1645	0.0068	0.0015	0.0523	0.0565	0.0071	0.1645	0.6322
	6	0.0024	-0.0000	0.0003	-0.1939	-0.02	0.0523	0.0068	0.0068	0.0068	0.0015	0.0523	0.0565	0.0071	0.1645	0.6322
	9	0.0000	0.0001	-0.0000	0.3948	-0.4241	0.0565	0.0015	0.0015	0.0071	0.0015	0.0565	0.0565	0.0071	0.1645	0.6322
304	2	0.600	0.05	-0.00	-0.0460	-0.02	15.11	15.09	0.1434	0.0089	0.0020	0.0736	0.0760	0.0089	0.1434	0.6107
	6	0.0025	-0.0000	0.0003	-0.0460	-0.02	0.0736	0.0089	0.0089	0.0089	0.0020	0.0736	0.0760	0.0089	0.1434	0.6107
	9	-0.0003	0.0001	-0.0000	-1.8135	0.9020	0.0760	0.0020	0.0020	0.0089	0.0020	0.0760	0.0760	0.0089	0.1434	0.6107
304	3	0.600	1.08	0.00	-0.0551	-0.02	15.12	15.10	0.1433	0.0095	0.0019	0.0801	0.0796	0.0090	0.1433	0.5987
	6	0.0029	-0.0000	0.0003	-0.0551	-0.02	0.0801	0.0095	0.0095	0.0095	0.0019	0.0801	0.0796	0.0090	0.1433	0.5987
	9	0.0001	0.0001	0.0000	5.7633	0.3663	0.0796	0.0019	0.0019	0.0090	0.0019	0.0796	0.0796	0.0090	0.1433	0.5987
304	4	0.600	3.15	-0.00	-0.0867	-0.02	15.11	15.09	0.1198	0.0095	0.0014	0.0831	0.0857	0.0095	0.1198	0.5571
	6	0.0022	-0.0000	0.0003	-0.0867	-0.02	0.0831	0.0095	0.0095	0.0095	0.0014	0.0831	0.0857	0.0095	0.1198	0.5571
	9	-0.0003	0.0001	-0.0000	-2.6694	0.0518	0.0857	0.0014	0.0014	0.0095	0.0014	0.0857	0.0857	0.0095	0.1198	0.5571
304	5	0.600	6.24	0.00	-0.2431	-0.02	15.10	15.08	0.0618	0.0094	0.0008	0.0832	0.0847	0.0092	0.0618	0.5482
	6	0.0016	-0.0000	0.0002	-0.2431	-0.02	0.0832	0.0094	0.0094	0.0094	0.0008	0.0832	0.0847	0.0092	0.0618	0.5482
	9	0.0011	0.0000	0.0002	1.0831	0.9336	0.0847	0.0008	0.0008	0.0092	0.0008	0.0847	0.0847	0.0092	0.0618	0.5482
304	6	0.601	9.29	0.00	-0.1809	-0.02	15.09	15.07	0.0463	0.0094	0.0003	0.0857	0.0853	0.0094	0.0463	0.5535
	6	0.0021	-0.0000	0.0003	-0.1809	-0.02	0.0857	0.0094	0.0094	0.0094	0.0003	0.0857	0.0853	0.0094	0.0463	0.5535
	9	0.0017	0.0002	0.0002	0.7265	0.7809	0.0853	0.0003	0.0003	0.0094	0.0003	0.0853	0.0853	0.0094	0.0463	0.5535
304	7	0.601	12.47	0.00	-0.2898	-0.02	15.09	15.07	0.0345	0.0102	0.0002	0.0907	0.0916	0.0101	0.0345	0.5521
	6	0.0017	-0.0001	0.0002	-0.2898	-0.02	0.0907	0.0102	0.0102	0.0102	0.0002	0.0907	0.0916	0.0101	0.0345	0.5521
	9	0.0008	0.0002	0.0001	1.2013	0.9606	0.0916	0.0002	0.0002	0.0101	0.0002	0.0916	0.0916	0.0101	0.0345	0.5521
304	8	0.600	0.03	-0.00	-0.0658	-0.02	15.11	15.09	0.1423	0.0088	0.0020	0.0736	0.0758	0.0088	0.1423	0.6116
	6	0.0029	-0.0000	0.0004	-0.0658	-0.02	0.0736	0.0088	0.0088	0.0088	0.0020	0.0736	0.0758	0.0088	0.1423	0.6116
	9	-0.0003	0.0001	0.0000	-2.6005	-0.2271	0.0758	0.0020	0.0020	0.0088	0.0020	0.0758	0.0758	0.0088	0.1423	0.6116
305	1	0.601	-0.03	0.00	0.3567	-2.01	-0.05	0.06	0.5489	0.0000	0.0001	0.0003	0.0005	0.0000	0.5489	0.5980
	6	0.0050	-0.0003	-0.0006	0.3567	-2.01	0.0003	0.06	0.5489	0.0000	0.0001	0.0003	0.0005	0.0000	0.5489	0.5980
	9	0.0003	0.0001	0.0000	2.2577	0.4668	0.0005	-0.0001	-0.0001	0.0000	-0.0001	0.0005	-0.0001	0.0000	2.2577	0.5980

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key															
305	2	7	0.601	-0.001	-0.003	0.000	0.000	1.7547	-1.003	-0.002	0.07	-0.02	0.3190					
		9	-0.0005	0.0001	0.0001	-0.0000	0.0000	1.4413	0.2654	0.0002	0.0001	0.0000	0.3674					
305	3	7	0.602	-0.001	-0.003	0.000	0.000	-0.3057	-0.53	-0.000	0.06	-0.02	0.1342					
		9	0.0016	0.0001	0.0001	0.0000	0.0000	2.5219	0.6386	0.0000	0.0001	0.0000	0.7403					
305	4	7	0.601	-0.001	-0.003	-0.000	-0.000	-0.0655	-0.002	-0.005	0.06	-0.01	0.7136					
		9	0.0030	0.0001	0.0001	0.0000	0.0000	4.7066	0.6353	0.0000	0.0001	0.0000	0.6143					
305	5	7	0.601	-0.001	-0.003	-0.000	-0.000	0.0262	0.49	-0.002	0.06	-0.02	0.4877					
		9	0.0050	0.0001	0.0001	0.0000	0.0000	-1.5136	0.6521	0.0004	0.0001	0.0000	0.5006					
305	6	7	0.600	-0.001	-0.003	-0.000	-0.000	0.0811	1.0024	-0.004	0.07	-0.02	0.5760					
		9	0.0083	0.0001	0.0001	0.0010	0.0001	0.7360	0.4876	0.0006	0.0001	0.0001	0.6057					
305	7	7	0.601	-0.001	-0.003	-0.000	-0.000	0.1152	2.07	-0.005	0.07	-0.02	0.8123					
		9	0.0131	0.0004	0.0001	0.0017	0.0000	2.2384	0.6460	0.0005	0.0001	0.0000	0.6473					
306	1	7	0.601	-0.001	-0.003	45.00	0.000	0.3051	-2.101	-0.0019	0.06	-0.02	0.7011					
		9	-0.0073	0.0000	0.0002	-0.0009	0.0000	43.8526	0.6117	0.0011	0.0001	0.0001	0.4539					
306	2	7	0.601	-0.001	-0.003	45.00	0.000	0.5211	-1.203	-0.0019	0.06	-0.02	0.7752					
		9	-0.0025	0.0002	0.0002	-0.0002	0.0000	-5.1183	0.9578	0.0009	0.0001	0.0000	0.4536					
306	3	7	0.601	-0.001	-0.003	44.99	0.000	2.5307	-0.53	-0.0023	0.06	-0.01	0.7059					
		9	-0.0002	0.0002	0.0002	0.0001	0.0001	1.1651	0.7788	0.0009	0.0001	0.0000	0.2708					
306	4	7	0.603	-0.001	-0.003	44.99	0.000	-0.3383	-0.002	-0.0024	0.06	-0.02	0.7588					
		9	0.0015	0.0009	0.0002	0.0001	0.0001	1.1758	0.9938	0.0003	0.0001	0.0000	0.0653					
306	5	7	0.602	-0.001	-0.003	44.99	0.000	-0.0130	0.47	-0.006	0.06	-0.02	0.7702					
		9	0.0034	0.0003	0.0002	0.0005	0.0001	3.5902	0.7753	-0.0021	0.0000	0.0000	0.3402					

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key															
306	6	7	0.601	-0.004	44.99	0.0669	0.7625	-0.0004	0.06	-0.0001	-0.0000	-1.5405	0.8647					
	6	9	0.0051	0.0002	0.0007	4.9064	2.4295	0.0014	0.0001	0.0001	0.0001	-0.3593	0.4930					
	7	6	0.601	-0.004	44.99	0.1060	2.07	-0.0009	0.07	-0.0001	-0.0001	-0.7620	0.5722					
	6	9	0.0104	0.0002	0.0014	5.8474	3.2839	0.0011	0.0000	0.0001	0.0001	-0.4020	0.5923					
	3	7	1.249	-3.10	0.00	-0.6250	-0.03	-0.09	0.06	-0.04	-0.0395	0.6890						
	6	9	0.0011	0.0002	0.0002	1.1397	0.5964	-0.0252	0.0000	-0.0035	0.0158	0.6982						
	4	7	1.251	-0.01	0.00	-0.4126	-0.02	-0.06	0.06	-0.02	-3.0905	1.3286						
	6	9	0.0020	0.0003	0.0002	1.2485	0.6323	-0.0003	0.0001	-0.0000	2.6473	0.7102						
	5	7	1.251	1.01	0.00	-0.4133	-0.02	-0.05	0.06	-0.01	0.1195	0.6705						
	6	9	0.0017	0.0003	0.0002	1.1975	0.6961	0.0090	0.0002	0.0012	-0.0920	0.6448						
	6	7	1.250	3.13	0.00	-0.6022	-0.02	-0.03	0.06	-0.01	0.0190	0.6565						
	6	9	0.0015	0.0003	0.0001	0.9960	0.7243	0.0267	0.0002	0.0035	-0.0418	0.6819						
	7	7	1.251	6.24	0.00	-1.8121	-0.02	-0.01	0.06	0.00	-0.0231	0.6695						
	6	9	0.0005	0.0003	0.0002	1.5399	0.7195	0.0521	0.0002	0.0069	-0.0526	0.6659						
	8	7	1.249	9.37	0.00	-2.9324	-0.02	0.01	0.06	0.01	-0.0338	0.6522						
	6	9	0.0003	0.0003	0.0002	0.9406	0.9298	0.0770	0.0008	0.0100	-0.0543	0.6543						
	9	7	1.249	12.62	0.00	-1.3419	-0.02	0.02	0.06	0.02	-0.0349	0.6345						
	6	9	0.0006	0.0003	0.0001	4.1625	1.5259	0.0985	0.0011	0.0124	-0.0598	0.6322						
	10	7	1.251	-0.03	0.00	-0.4557	-0.02	-0.06	0.05	-0.03	-1.8690	1.8026						
	6	9	0.0020	0.0003	0.0002	2.0543	0.5552	-0.0004	0.0001	-0.0001	5.4922	3.1171						
	1	7	1.250	-0.01	45.00	-6.8324	-0.02	-0.07	0.06	-0.02	-0.7766	1.3908						
	6	9	0.0003	0.0003	0.0001	4.8242	1.6318	-0.0002	0.0001	-0.0001	4.5004	3.1860						

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key													
309	2	7	1.251	-5.12	45.00	-0.0518	-0.6630	-0.0177	0.0001	0.03	-0.04	-0.0405	0.7228		
		6	-0.0183	-0.0001	-0.0024	0.0792	0.6661	-0.0179	-0.0001	-0.0025	-0.0025	0.0495	0.7111		
		9	-0.0156	0.0002	-0.0020	-0.0792	0.6661	-0.0179	-0.0001	-0.0025	-0.0025	0.0495	0.7111		
309	3	7	1.250	-0.00	45.00	-2.5354	-0.7188	-0.0006	0.0001	0.06	-0.03	-0.8284	2.5792		
		6	0.0003	-0.0001	0.0001	2.3499	0.9481	-0.0006	-0.0001	-0.0001	-0.0001	1.1787	1.0727		
		9	0.0007	0.0003	0.0001	-2.3499	0.9481	-0.0006	-0.0001	-0.0001	-0.0001	1.1787	1.0727		
309	4	7	1.249	1.00	45.00	-0.1030	-0.6656	-0.0041	0.0002	0.07	-0.01	0.2506	0.6833		
		6	0.0062	-0.0001	0.0008	0.2861	0.6788	0.0053	-0.0001	0.0002	0.0005	-0.1528	0.6059		
		9	0.0066	0.0003	0.0009	0.2861	0.6788	0.0053	-0.0001	0.0002	0.0006	-0.1528	0.6059		
309	5	7	1.250	5.14	44.99	-0.0357	-0.6666	-0.0160	0.0000	0.08	-0.01	0.0286	0.6131		
		6	0.0188	-0.0001	0.0025	0.0817	0.6942	0.0175	-0.0002	0.0000	0.0019	0.0286	0.6590		
		9	0.0195	0.0003	0.0027	0.0817	0.6942	0.0175	-0.0002	0.0000	0.0023	-0.0699	0.6590		
309	6	7	1.250	6.26	45.00	-0.0417	0.6817	-0.0346	-0.0000	0.12	0.00	-0.0127	0.6603		
		6	0.0369	-0.0003	0.0055	0.0117	0.6868	0.0372	-0.0004	0.0000	0.0045	-0.0661	0.6642		
		9	0.0401	0.0000	0.0055	0.0117	0.6868	0.0372	-0.0004	0.0000	0.0049	-0.0661	0.6642		
309	7	7	1.250	9.41	44.99	-0.0485	0.6801	-0.0513	-0.0002	0.13	0.01	-0.0285	0.6623		
		6	0.0535	-0.0005	0.0072	0.0125	0.6732	0.0557	-0.0005	0.0000	0.0068	-0.0536	0.6639		
		9	0.0579	-0.0001	0.0078	0.0125	0.6732	0.0557	-0.0005	0.0000	0.0074	-0.0536	0.6639		
309	8	7	1.249	12.57	44.99	-0.0545	0.6712	-0.0672	-0.0005	0.15	0.02	-0.0395	0.6682		
		6	0.0687	-0.0007	0.0092	0.0205	0.6596	0.0751	-0.0007	0.0000	0.0089	-0.0510	0.6496		
		9	0.0764	-0.0003	0.0100	0.0205	0.6596	0.0751	-0.0007	0.0000	0.0097	-0.0510	0.6496		
309	9	7	1.249	-0.01	45.00	-4.2747	-0.03	-0.0006	0.0001	0.06	-0.02	-1.1753	0.9535		
		6	0.0002	-0.0001	0.0000	3.0502	1.2164	-0.0000	-0.0002	0.0001	-0.0001	0.0001	0.9535		
		9	0.0006	0.0003	0.0001	-3.0502	1.1180	0.0000	-0.0002	-0.0001	-0.0001	0.0001	0.9535		
310	1	7	1.250	-5.06	-0.00	-0.4805	-0.4989	0.0047	0.0000	0.05	0.00	0.0812	0.6721		
		6	0.0018	-0.0001	0.0001	1.4769	0.3754	0.0049	-0.0002	0.0000	0.0007	-0.2210	0.7283		
		9	0.0011	0.0003	0.0000	1.4769	0.3754	0.0049	-0.0002	0.0000	0.0007	-0.2210	0.7283		
310	2	7	1.248	0.01	0.00	-0.4166	-0.402	0.0317	-0.0000	0.05	0.02	-0.0113	0.6793		
		6	0.0021	-0.0001	0.0002	1.9127	0.5464	0.0307	-0.0004	0.0000	0.0043	-0.0668	0.7005		
		9	0.0010	0.0003	0.0000	1.9127	0.5464	0.0307	-0.0004	0.0000	0.0043	-0.0668	0.7005		
310	3	7	1.248	1.05	0.00	-0.4297	-0.4955	0.0417	-0.0001	0.05	0.03	-0.0210	0.6852		
		6	0.0019	-0.0001	0.0001	1.8336	0.4696	0.0402	-0.0005	0.0001	0.0055	-0.0635	0.6906		
		9	0.0010	0.0003	0.0000	1.8336	0.4696	0.0402	-0.0005	0.0001	0.0055	-0.0635	0.6906		

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key													
310	4	1.249	5.17	0.00	-0.4961	-0.02	6.06	0.0593	0.0003	6.04	-0.0314	0.0077	0.0003	0.0079	0.6733
	6	0.0014	-0.0001	0.0001	1.6311	0.3850	0.0573	-0.0006	0.0077	-0.0591	0.0079	0.0003	0.0006	0.0079	0.6779
	9	0.0011	0.0003	0.0001	0.0001	0.6215	0.0573	-0.0006	0.0077	-0.0591	0.0079	0.0003	0.0006	0.0079	0.6779
310	5	1.250	6.29	0.00	-1.3513	-0.02	6.08	0.0006	6.06	-0.0402	0.0107	0.0006	0.0009	0.6501	
	6	0.0007	-0.0002	0.0000	1.1394	0.4969	0.0830	-0.0006	0.0107	-0.0594	0.0107	0.0006	0.0009	0.6515	
	9	0.0019	0.0004	0.0002	0.0002	0.7730	0.0827	-0.0009	0.0107	-0.0594	0.0107	0.0006	0.0009	0.6515	
310	6	1.249	9.43	0.00	-3.7686	-0.02	6.10	0.0007	6.06	-0.0357	0.0129	0.0007	0.0012	0.6351	
	6	0.0002	-0.0002	0.0000	1.2938	0.6469	0.1021	-0.0007	0.0129	-0.0602	0.0129	0.0007	0.0012	0.6271	
	9	0.0015	0.0004	0.0002	0.0002	0.8400	0.1034	-0.0012	0.0129	-0.0602	0.0129	0.0007	0.0012	0.6271	
310	7	1.250	12.63	0.00	-5.9125	-0.02	6.11	0.0010	6.07	-0.0454	0.0148	0.0010	0.0015	0.6175	
	8	0.0007	-0.0001	0.0000	1.2959	0.5827	0.1202	-0.0010	0.0148	-0.0660	0.0148	0.0010	0.0015	0.6151	
	9	0.0003	0.0004	0.0001	0.0001	1.6488	0.1203	-0.0015	0.0148	-0.0660	0.0148	0.0010	0.0015	0.6151	
310	8	1.249	-0.00	-0.00	-0.4748	-0.02	6.03	0.0001	6.02	-0.0159	0.0042	0.0001	0.0004	0.6725	
	8	0.0020	-0.0001	0.0002	2.5258	0.4992	0.0316	-0.0001	0.0042	-0.0710	0.0042	0.0001	0.0004	0.6925	
	9	0.0008	0.0004	0.0001	0.0001	0.7463	0.0306	-0.0004	0.0042	-0.0710	0.0042	0.0001	0.0004	0.6925	
311	3	1.247	-3.06	45.00	-0.0788	6.03	6.01	6.05	6.00	0.0833	0.0018	6.00	6.05	0.7418	
	6	0.0115	-0.0001	0.0017	0.0879	0.7417	0.0121	0.0002	0.0018	0.1189	0.0018	0.0002	0.0002	0.7110	
	9	0.0117	0.0002	0.0017	0.0879	0.7400	0.0121	-0.0002	0.0018	0.1189	0.0018	0.0002	0.0002	0.7110	
311	4	1.248	0.02	45.00	-0.0516	6.05	6.04	6.07	6.01	-0.0060	0.0041	6.01	6.07	0.6871	
	6	0.0308	-0.0003	0.0044	-0.0013	0.7149	0.0302	-0.0000	0.0041	-0.0618	0.0041	0.0000	0.0003	0.6982	
	9	0.0313	-0.0000	0.0044	-0.0013	0.7092	0.0311	-0.0003	0.0041	-0.0618	0.0041	0.0000	0.0003	0.6982	
311	5	1.251	1.06	44.99	-0.0513	6.06	6.04	6.08	6.01	-0.0118	0.0053	6.01	6.08	0.6977	
	6	0.0374	-0.0003	0.0053	-0.0098	0.7089	0.0356	-0.0000	0.0053	-0.0599	0.0053	0.0000	0.0004	0.7017	
	9	0.0385	-0.0000	0.0053	-0.0098	0.6995	0.0378	-0.0004	0.0053	-0.0599	0.0053	0.0000	0.0004	0.7017	
311	6	1.250	3.15	44.99	-0.0509	6.07	6.04	6.10	6.03	-0.0281	0.0065	6.03	6.10	0.6893	
	6	0.0503	-0.0005	0.0069	-0.0195	0.6938	0.0472	-0.0002	0.0065	-0.0623	0.0065	0.0002	0.0006	0.6785	
	9	0.0521	-0.0002	0.0071	-0.0195	0.6872	0.0513	-0.0006	0.0065	-0.0623	0.0065	0.0002	0.0006	0.6785	
311	7	1.247	6.30	44.99	-0.0587	6.08	6.06	6.12	6.03	-0.0397	0.0088	6.03	6.12	0.6776	
	6	0.0670	-0.0007	0.0091	-0.0258	0.6795	0.0649	-0.0005	0.0088	-0.0551	0.0088	0.0005	0.0007	0.6668	
	9	0.0715	-0.0003	0.0095	-0.0258	0.6690	0.0696	-0.0007	0.0088	-0.0551	0.0088	0.0005	0.0007	0.6668	
311	8	1.249	9.44	44.99	-0.0612	6.10	6.08	6.14	6.05	-0.0458	0.0105	6.05	6.14	0.6622	
	8	0.0813	-0.0009	0.0108	-0.0263	0.6685	0.0799	-0.0009	0.0105	-0.0538	0.0105	0.0007	0.0009	0.6484	
	9	0.0873	-0.0004	0.0114	-0.0263	0.6550	0.0878	-0.0009	0.0105	-0.0538	0.0105	0.0007	0.0009	0.6484	

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																	
311	9	1.250	12.60	44.99	-0.0595	6.11	6.09	6.15	6.06	-0.0494	0.6467	0.0927	-0.0009	0.0120	-0.0521	0.6312	0.0120	-0.0494	0.6467
	6	0.0927	-0.0011	0.0121	-0.0287	0.6541	0.0927	-0.0009	0.0120	-0.0521	0.6312	0.0927	-0.0009	0.0120	-0.0521	0.6312	0.0120	-0.0494	0.6467
	9	0.1031	-0.0005	0.0131	-0.0287	0.6361	0.1045	-0.0010	0.0132	-0.0521	0.6312	0.1045	-0.0010	0.0132	-0.0521	0.6312	0.0132	-0.0494	0.6467
311	10	1.248	0.05	45.00	-0.0536	6.04	6.03	6.08	6.00	-0.0089	0.6848	0.0303	-0.0000	0.0041	-0.0635	0.6795	0.0041	-0.0089	0.6848
	6	0.0308	-0.0003	0.0043	-0.0083	0.7059	0.0303	-0.0000	0.0041	-0.0635	0.6795	0.0303	-0.0000	0.0041	-0.0635	0.6795	0.0041	-0.0089	0.6848
	9	0.0313	0.0000	0.0044	0.0083	0.7139	0.0313	-0.0003	0.0042	-0.0635	0.6795	0.0313	-0.0003	0.0042	-0.0635	0.6795	0.0042	-0.0089	0.6848
312	1	1.249	-2.94	45.00	-0.0579	15.08	15.05	15.05	15.02	-0.0343	0.6913	0.0605	-0.0004	0.0083	-0.0811	0.6866	0.0083	-0.0343	0.6913
	6	0.0600	-0.0006	0.0083	-0.0410	0.6925	0.0605	-0.0004	0.0083	-0.0811	0.6866	0.0605	-0.0004	0.0083	-0.0811	0.6866	0.0083	-0.0343	0.6913
	9	0.0582	-0.0004	0.0081	-0.0410	0.6967	0.0590	-0.0009	0.0081	-0.0811	0.6866	0.0590	-0.0009	0.0081	-0.0811	0.6866	0.0081	-0.0343	0.6913
312	2	1.248	0.12	44.99	-0.0578	15.10	15.07	15.07	15.03	-0.0419	0.6749	0.0761	-0.0006	0.0102	-0.0880	0.6659	0.0102	-0.0419	0.6749
	6	0.0770	-0.0008	0.0104	-0.0379	0.6767	0.0761	-0.0006	0.0102	-0.0880	0.6659	0.0761	-0.0006	0.0102	-0.0880	0.6659	0.0102	-0.0419	0.6749
	9	0.0766	-0.0005	0.0103	-0.0379	0.6752	0.0774	-0.0010	0.0103	-0.0880	0.6659	0.0774	-0.0010	0.0103	-0.0880	0.6659	0.0103	-0.0419	0.6749
312	3	1.248	1.20	44.99	-0.0577	15.10	15.07	15.08	15.04	-0.0425	0.6673	0.0814	-0.0006	0.0108	-0.0656	0.6613	0.0108	-0.0425	0.6673
	6	0.0829	-0.0009	0.0110	-0.0352	0.6669	0.0814	-0.0006	0.0108	-0.0656	0.6613	0.0814	-0.0006	0.0108	-0.0656	0.6613	0.0108	-0.0425	0.6673
	9	0.0830	-0.0005	0.0110	-0.0352	0.6655	0.0838	-0.0011	0.0110	-0.0656	0.6613	0.0838	-0.0011	0.0110	-0.0656	0.6613	0.0110	-0.0425	0.6673
312	4	1.248	3.28	44.99	-0.0582	15.12	15.08	15.10	15.05	-0.0442	0.6576	0.0903	-0.0007	0.0118	-0.0629	0.6437	0.0118	-0.0442	0.6576
	6	0.0923	-0.0010	0.0121	-0.0311	0.6583	0.0903	-0.0007	0.0118	-0.0629	0.6437	0.0903	-0.0007	0.0118	-0.0629	0.6437	0.0118	-0.0442	0.6576
	9	0.0938	-0.0005	0.0121	-0.0311	0.6496	0.0952	-0.0011	0.0122	-0.0629	0.6437	0.0952	-0.0011	0.0122	-0.0629	0.6437	0.0122	-0.0442	0.6576
312	5	1.248	6.37	44.99	-0.0606	15.13	15.09	15.11	15.06	-0.0472	0.6436	0.1030	-0.0009	0.0132	-0.0589	0.6269	0.0132	-0.0472	0.6436
	6	0.1046	-0.0012	0.0133	-0.0344	0.6383	0.1030	-0.0009	0.0132	-0.0589	0.6269	0.1030	-0.0009	0.0132	-0.0589	0.6269	0.0132	-0.0472	0.6436
	9	0.1104	-0.0007	0.0139	-0.0344	0.6302	0.1109	-0.0013	0.0139	-0.0589	0.6269	0.1109	-0.0013	0.0139	-0.0589	0.6269	0.0139	-0.0472	0.6436
312	6	1.248	9.52	44.99	-0.0673	15.13	15.09	15.13	15.07	-0.0535	0.6354	0.1143	-0.0013	0.0145	-0.0523	0.6108	0.0145	-0.0535	0.6354
	6	0.1140	-0.0015	0.0144	-0.0354	0.6313	0.1143	-0.0013	0.0145	-0.0523	0.6108	0.1143	-0.0013	0.0145	-0.0523	0.6108	0.0145	-0.0535	0.6354
	9	0.1234	-0.0008	0.0153	-0.0354	0.6200	0.1246	-0.0013	0.0152	-0.0523	0.6108	0.1246	-0.0013	0.0152	-0.0523	0.6108	0.0152	-0.0535	0.6354
312	7	1.250	12.67	44.99	-0.0668	15.14	15.10	15.14	15.08	-0.0609	0.6286	0.1213	-0.0014	0.0152	-0.0382	0.5934	0.0152	-0.0609	0.6286
	6	0.1217	-0.0016	0.0152	-0.0233	0.6272	0.1213	-0.0014	0.0152	-0.0382	0.5934	0.1213	-0.0014	0.0152	-0.0382	0.5934	0.0152	-0.0609	0.6286
	9	0.1293	-0.0006	0.0156	-0.0233	0.6041	0.1305	-0.0009	0.0154	-0.0382	0.5934	0.1305	-0.0009	0.0154	-0.0382	0.5934	0.0154	-0.0609	0.6286
312	8	1.248	0.14	44.99	-0.0589	15.10	15.07	15.07	15.03	-0.0438	0.6703	0.0766	-0.0006	0.0102	-0.0661	0.6625	0.0102	-0.0438	0.6703
	6	0.0775	-0.0009	0.0104	-0.0355	0.6721	0.0766	-0.0006	0.0102	-0.0661	0.6625	0.0766	-0.0006	0.0102	-0.0661	0.6625	0.0102	-0.0438	0.6703
	9	0.0769	-0.0005	0.0103	-0.0355	0.6749	0.0776	-0.0010	0.0102	-0.0661	0.6625	0.0776	-0.0010	0.0102	-0.0661	0.6625	0.0102	-0.0438	0.6703
313	9	1.249	-2.99	0.00	-0.8263	-0.04	15.06	0.06	15.01	-0.0384	0.7012	0.0512	-0.0003	0.0073	-0.0732	0.6967	0.0073	-0.0384	0.7012
	6	0.0009	-0.0001	0.0001	-0.0903	0.9769	0.0512	-0.0003	0.0073	-0.0732	0.6967	0.0512	-0.0003	0.0073	-0.0732	0.6967	0.0073	-0.0384	0.7012
	9	0.0012	0.0002	0.0000	1.0903	0.3238	0.0525	-0.0007	0.0073	-0.0732	0.6967	0.0525	-0.0007	0.0073	-0.0732	0.6967	0.0073	-0.0384	0.7012

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

← See Key →

Run Pt.	Cd																						
313	10	7	9	1.248	-0.0018	0.0013	0.06	0.0002	-0.4251	1.1416	-0.04	0.7160	0.5892	15.07	0.0774	0.0778	0.07	0.0006	0.0104	-0.0405	0.0103	0.6732	0.6657
313	11	7	9	1.248	-0.0017	0.0011	1.11	0.0002	-0.4128	1.4222	-0.04	0.7562	0.9067	15.08	0.0857	0.0859	0.07	0.0006	0.0113	-0.0378	0.0112	0.6637	0.6558
313	12	7	9	1.250	-0.0008	0.0014	3.23	0.0001	-0.7399	1.1784	-0.04	1.0063	0.6905	15.10	0.0995	0.0993	0.07	0.0007	0.0128	-0.0373	0.0127	0.6462	0.6411
313	13	7	9	1.248	-0.0005	0.0022	6.33	0.0001	-1.7904	0.8584	-0.04	1.0085	0.6261	15.11	0.1162	0.1170	0.07	0.0009	0.0145	-0.0398	0.0145	0.6312	0.6201
313	14	7	9	1.248	-0.0002	0.0018	9.47	0.0002	-4.6176	0.8626	-0.04	2.2617	0.7581	15.12	0.1310	0.1315	0.07	0.0012	0.0161	-0.0463	0.0160	0.6176	0.6085
313	15	7	9	1.248	-0.0006	0.0010	12.68	0.0001	-1.1656	1.9394	-0.04	1.0210	0.8402	15.13	0.1340	0.1343	0.08	0.0011	0.0161	-0.0413	0.0160	0.6017	0.5979
313	16	7	9	1.249	-0.0018	0.0010	0.09	0.0002	-0.4462	1.7790	-0.04	0.6536	0.7865	15.07	0.0777	0.0777	0.06	0.0006	0.0103	-0.0431	0.0103	0.6685	0.6623
314	1	7	9	0.901	-0.0023	0.0000	-2.98	0.0003	-0.2122	29.7932	-0.03	0.8430	5.2922	15.07	0.0591	0.0609	0.06	0.0010	0.0075	0.0915	0.0077	0.6370	0.6358
314	2	7	9	0.900	-0.0027	0.0006	0.06	0.0004	-0.0598	-2.4103	-0.03	0.7749	0.3455	15.09	0.0798	0.0803	0.06	0.0011	0.0096	0.0717	0.0095	0.6028	0.5916
314	3	7	9	0.897	-0.0026	0.0003	1.09	0.0004	0.0058	2.5777	-0.03	0.7964	0.8907	15.08	0.0814	0.0830	0.06	0.0011	0.0095	0.0733	0.0096	0.5869	0.5782
314	4	7	9	0.899	-0.0022	0.0005	3.17	0.0004	-0.0424	-2.8343	-0.03	0.8934	1.5083	15.08	0.0863	0.0888	0.06	0.0006	0.0101	0.0396	0.0101	0.5853	0.5720

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Pt.	Cd	See Key															
314	5	7	0.898	6.22	0.003	0.00	-0.2022	-0.02	15.08	0.953	0.06	15.05	0.0164	0.5797				
	6	6	0.0019	-0.0000	0.0002	0.0003	1.3564	0.8457	0.0954	0.0000	0.0003	0.0110	-0.0008	0.5708				
	9	9	0.0013	0.0003	0.0002	0.0002	0.0002	0.8452	0.0954	-0.0000	0.0000	0.0108	0.0008	0.5708				
314	6	7	0.897	9.29	0.003	0.00	-0.2547	-0.02	15.08	0.953	0.06	15.05	0.0015	0.5769				
	6	6	0.0018	-0.0000	0.0003	0.0003	1.3896	0.9578	0.1038	0.0000	0.0000	0.0119	-0.0096	0.5718				
	9	9	0.0012	0.0003	0.0003	0.0003	1.2733	1.2733	0.1007	-0.0001	0.0001	0.0115	-0.0096	0.5718				
314	7	7	0.900	12.45	0.00	0.00	-0.3409	-0.02	15.08	0.953	0.06	15.05	0.0008	0.5789				
	6	6	0.0021	-0.0001	0.0002	0.0002	2.3799	0.5861	0.1047	0.0000	0.0000	0.0121	-0.0159	0.5699				
	9	9	0.0007	0.0003	0.0002	0.0002	1.4314	1.4314	0.1047	-0.0003	0.0003	0.0119	-0.0159	0.5699				
314	8	7	0.898	0.06	-0.00	0.00	0.0024	-0.03	15.07	0.953	0.06	15.05	0.0706	0.5962				
	6	6	0.0023	0.0000	0.0004	0.0004	2.6789	0.9815	0.0799	0.0011	0.0011	0.0095	0.0429	0.5892				
	9	9	0.0005	0.0002	0.0000	0.0000	0.7452	0.7452	0.0801	0.0006	0.0006	0.0094	0.0429	0.5892				
315	1	7	0.899	-2.95	44.99	44.99	0.0428	15.09	15.07	15.07	15.07	15.05	0.0753	0.6312				
	6	6	0.0695	0.0005	0.0087	0.0087	0.0876	0.6282	0.0686	0.0010	0.0010	0.0086	0.0485	0.6325				
	9	9	0.0646	0.0011	0.0081	0.0081	0.0876	0.6311	0.0666	0.0006	0.0006	0.0084	0.0485	0.6325				
315	2	7	0.898	0.14	44.99	44.99	0.0408	15.11	15.08	15.08	15.08	15.05	0.0539	0.6099				
	6	6	0.0838	0.0006	0.0101	0.0101	0.0812	0.6020	0.0836	0.0009	0.0009	0.0102	0.0493	0.5998				
	9	9	0.0786	0.0012	0.0094	0.0094	0.0812	0.5987	0.0805	0.0007	0.0007	0.0096	0.0493	0.5998				
315	3	7	0.898	1.17	44.99	44.99	0.0410	15.11	15.08	15.08	15.08	15.06	0.0519	0.6059				
	6	6	0.0874	0.0007	0.0106	0.0106	0.0815	0.6091	0.0873	0.0009	0.0009	0.0105	0.0542	0.5891				
	9	9	0.0814	0.0013	0.0096	0.0096	0.0815	0.5934	0.0836	0.0009	0.0009	0.0098	0.0542	0.5891				
315	4	7	0.897	3.20	44.99	44.99	0.0381	15.12	15.08	15.08	15.08	15.06	0.0453	0.5972				
	6	6	0.0928	0.0007	0.0110	0.0110	0.0554	0.5929	0.0934	0.0008	0.0008	0.0111	0.0407	0.5841				
	9	9	0.0866	0.0009	0.0101	0.0101	0.0554	0.5844	0.0898	0.0007	0.0007	0.0105	0.0407	0.5841				
315	5	7	0.898	6.27	44.99	44.99	0.0124	15.12	15.09	15.09	15.09	15.05	0.0223	0.5834				
	6	6	0.0994	0.0002	0.0117	0.0117	0.0284	0.5922	0.1011	0.0004	0.0004	0.0118	0.0129	0.5769				
	9	9	0.0935	0.0005	0.0109	0.0109	0.0284	0.5860	0.0964	0.0002	0.0002	0.0111	0.0129	0.5769				
315	6	7	0.898	9.35	45.00	45.00	-0.0027	15.12	15.09	15.09	15.09	15.06	0.0084	0.5852				
	6	6	0.1047	-0.0000	0.0119	0.0119	0.0154	0.5719	0.1049	0.0001	0.0001	0.0122	0.0084	0.5852				
	9	9	0.1020	0.0003	0.0118	0.0118	0.0154	0.5814	0.1018	0.0002	0.0002	0.0116	0.0084	0.5852				
315	7	7	0.897	12.47	45.00	45.00	-0.0121	15.12	15.09	15.09	15.09	15.06	-0.0058	0.5799				
	6	6	0.1074	-0.0002	0.0123	0.0123	0.0176	0.5751	0.1122	-0.0001	-0.0001	0.0130	-0.0058	0.5799				
	9	9	0.1033	0.0003	0.0119	0.0119	0.0176	0.5774	0.1054	0.0001	0.0001	0.0121	-0.0058	0.5799				

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																	
315	8	0.898	0.14	44.99	15.11	15.08	15.09	15.05	0.0576	0.6128	0.0831	0.0006	0.0102	0.0009	0.0007	0.0009	0.0102	0.0495	0.5971
	8	0.0783	0.0012	0.0095	0.0414	0.0855	0.0007	0.0096											
	9				0.0827	0.0806													
316	1	0.899	-3.04	44.99	6.03	6.02	6.06	6.01	0.1859	0.6996	0.0163	0.0001	0.0020	0.0005	0.0001	0.0005	0.0020	0.0333	0.6938
	6	0.0144	0.0006	0.0020	0.0490	0.0148	0.0001	0.0021											
	9				0.2310	0.0157													
316	2	0.899	0.05	44.99	6.06	6.04	6.09	6.03	0.0823	0.6729	0.0415	0.0003	0.0051	0.0006	0.0004	0.0006	0.0051	0.0636	0.6858
	6	0.0370	0.0010	0.0049	0.0379	0.0382	0.0004	0.0052											
	9				0.1374	0.0385													
316	3	0.898	1.09	44.99	6.07	6.05	6.10	6.03	0.0943	0.6594	0.0469	0.0006	0.0056	0.0008	0.0006	0.0008	0.0056	0.0704	0.6723
	6	0.0469	0.0006	0.0064	0.0639	0.0432	0.0006	0.0061											
	9	0.0431	0.0012	0.0056	0.1413	0.0454													
316	4	0.898	3.17	44.99	6.08	6.06	6.12	6.04	0.0993	0.6383	0.0568	0.0008	0.0068	0.0010	0.0009	0.0010	0.0068	0.0824	0.6385
	6	0.0568	0.0014	0.0071	0.0767	0.0534	0.0009	0.0072											
	9				0.1308	0.0565													
316	5	0.897	6.22	44.99	6.09	6.08	6.13	6.05	0.0837	0.6227	0.0727	0.0008	0.0087	0.0011	0.0010	0.0010	0.0087	0.0735	0.6045
	6	0.0724	0.0015	0.0086	0.0602	0.0703	0.0010	0.0086											
	9				0.1049	0.0712													
316	6	0.897	9.32	44.99	6.11	6.09	6.13	6.07	0.0585	0.5931	0.0845	0.0010	0.0096	0.0009	0.0010	0.0009	0.0096	0.0659	0.5764
	6	0.0805	0.0012	0.0093	0.0548	0.0844	0.0010	0.0096											
	9				0.0796	0.0833													
316	7	0.898	12.44	44.99	6.11	6.09	6.13	6.05	0.0452	0.5765	0.0930	0.0006	0.0106	0.0008	0.0005	0.0010	0.0106	0.0283	0.5755
	6	0.0867	0.0007	0.0101	0.0337	0.0926	0.0005	0.0102											
	9				0.0408	0.0889													
316	8	0.897	0.07	44.99	6.06	6.04	6.09	6.02	0.0830	0.6755	0.0414	0.0003	0.0051	0.0006	0.0004	0.0006	0.0051	0.0641	0.6879
	6	0.0377	0.0010	0.0049	0.0406	0.0383	0.0004	0.0053											
	9				0.1397	0.0386													
317	1	0.898	-3.04	-0.00	-0.2051	6.01	0.06	5.99	0.4228	0.7518	0.0027	0.0001	0.0004	0.0004	0.0000	0.0000	0.0004	0.0480	0.7245
	6	0.0002	0.0002	-0.0000	4.5327	0.0054	0.0000	0.0010											
	9				-0.0073	0.0074													
317	2	0.898	-0.00	-0.00	-0.0364	6.04	0.06	6.02	0.1144	0.6918	0.0029	0.0000	0.0008	0.0008	0.0003	0.0008	0.0052	0.0456	0.6918
	6	0.0008	0.0002	0.0000	1.6373	0.0389	0.0003	0.0055											
	9				-0.2462	0.0400	0.0000	0.0052											

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																		
317	3	7	0.897	1.03	-0.003	-0.0117	-0.03	6.06	0.0011	6.03	0.1222	0.6404	0.0030	-0.0000	0.0000	0.0471	0.0005	0.0060	0.0624	0.6672
317	4	7	0.898	3.16	0.003	-0.1212	-0.02	6.07	0.0012	6.05	0.0999	0.6136	0.0022	0.0000	0.0000	0.0647	0.0008	0.0079	0.0690	0.6272
317	5	7	0.898	6.19	0.003	-0.3621	-0.03	6.09	0.0013	6.07	0.0830	0.5858	0.0015	0.0002	0.0000	0.0801	0.0009	0.0094	0.0593	0.5778
317	6	7	0.898	9.27	0.002	-0.2900	-0.02	6.08	0.0005	6.06	0.0323	0.5762	0.0017	0.0002	0.0000	0.0890	0.0002	0.0102	0.0146	0.5704
317	7	7	0.896	12.44	0.001	-0.3641	-0.02	6.08	0.0001	6.06	0.0081	0.5738	0.0018	0.0001	0.0000	0.0989	0.0001	0.0113	0.0016	0.5703
317	8	7	0.899	0.02	-0.003	-0.0167	-0.04	6.04	0.0003	6.02	0.1125	0.6644	0.0029	0.0000	0.0000	0.0389	0.0003	0.0054	0.0500	0.6904
318	1	7	0.898	3.09	-0.003	-0.2074	-0.03	-0.10	0.06	-0.05	0.0710	0.6867	0.0025	0.0000	-0.0291	-0.0004	-0.0040	0.1407	0.6895	
318	2	7	0.898	-0.02	-0.004	-0.0378	-0.02	-0.06	0.06	-0.02	-26.3251	4.9630	0.0007	0.0000	-0.0014	-0.0002	-0.0001	-0.6952	0.6660	
318	3	7	0.898	1.01	-0.003	-0.0292	-0.03	-0.04	0.06	-0.01	0.3718	0.7302	0.0033	0.0000	0.0081	0.0001	0.0011	0.0584	0.6443	
318	4	7	0.896	3.10	0.004	-0.0592	-0.02	-0.03	0.06	0.00	0.1126	0.6527	0.0024	0.0000	0.0331	0.0002	0.0043	0.0481	0.6745	
318	5	7	0.899	6.16	0.002	-0.1661	-0.03	0.01	0.06	0.03	0.1236	0.6184	0.0022	0.0000	0.0573	0.0009	0.0070	0.0786	0.6213	

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																		
318	6	0.899	9.28	0.002	-0.2601	-0.003	0.0791	0.05	0.0092	0.0703	0.5072	0.0018	-0.0000	0.0002	-0.6958	0.0791	0.0012	0.0092	0.0549	0.5800
	6	0.0015	0.0003	0.0002	1.2095	0.7238	0.0807	0.0008	0.0093	0.0549	0.5800	0.0015	0.0003	0.0002	0.7238	0.0807	0.0008	0.0093	0.0549	0.5800
318	7	0.897	12.43	0.000	-0.3296	-0.002	0.0907	0.05	0.0103	0.0270	0.5697	0.0021	-0.0001	0.0001	0.4962	0.0911	0.0002	0.0104	0.0152	0.5706
	6	0.0014	0.0004	0.0001	1.4936	0.5886	0.0911	0.0004	0.0104	0.0270	0.5697	0.0014	0.0004	0.0001	0.4962	0.0911	0.0002	0.0104	0.0152	0.5706
318	8	0.898	-0.004	-0.000	-0.0560	-0.003	-0.004	0.06	-0.000	2.3046	-1.1259	0.0026	-0.000	0.0000	0.8819	0.0015	0.0001	-0.000	-0.6304	0.5443
	6	0.0006	0.0003	0.0000	2.6003	0.4883	0.0015	0.0001	-0.000	2.3046	-1.1259	0.0006	0.0003	0.0000	0.4883	0.0015	0.0001	-0.000	-0.6304	0.5443
319	1	0.899	-3.09	45.00	0.1976	0.6501	-0.0214	0.03	-0.003	0.0754	0.6950	-0.0193	-0.0007	-0.0023	0.6766	-0.0189	-0.0007	-0.0026	0.1967	0.7110
	6	0.0174	-0.0001	-0.0023	0.0393	0.6766	-0.0189	-0.0007	-0.0026	0.1967	0.7110	0.0174	-0.0001	-0.0023	0.6766	-0.0189	-0.0007	-0.0026	0.1967	0.7110
319	2	0.898	-0.001	44.99	-0.5887	-0.004	-0.0019	0.06	-0.000	-0.4280	0.7716	0.0010	-0.0003	0.0001	1.5236	-0.0019	0.0001	-0.000	-0.7366	0.4611
	6	0.0014	0.0003	0.0001	1.1967	0.4034	0.0019	0.0001	-0.000	-0.4280	0.7716	0.0014	0.0003	0.0001	1.5236	-0.0019	0.0001	-0.000	-0.7366	0.4611
319	3	0.898	1.02	44.99	0.1324	0.7360	-0.0032	0.07	-0.001	0.4921	0.4325	0.0066	0.0001	0.0002	0.7059	0.0032	0.0000	0.0007	0.386	0.5914
	6	0.0070	0.0005	0.0010	0.4015	0.7059	0.0032	0.0000	0.0007	0.4921	0.4325	0.0070	0.0005	0.0010	0.7059	0.0032	0.0000	0.0007	0.386	0.5914
319	4	0.898	3.10	44.99	0.0761	0.6845	-0.004	0.09	-0.000	0.2301	0.6639	0.0237	0.0003	0.0003	0.6705	0.0212	0.0002	0.0028	0.5476	0.7109
	6	0.0237	0.0008	0.0031	0.1734	0.6705	0.0212	0.0002	0.0028	0.2301	0.6639	0.0237	0.0003	0.0003	0.6705	0.0212	0.0002	0.0028	0.5476	0.7109
319	5	0.901	6.17	44.99	0.0925	0.6343	-0.001	0.12	0.000	0.1212	0.6740	0.0442	0.0008	0.0057	0.6343	0.0381	0.0005	0.0058	0.1212	0.6597
	6	0.0450	0.0013	0.0057	0.1484	0.6343	0.0381	0.0005	0.0058	0.1212	0.6597	0.0450	0.0013	0.0057	0.6343	0.0381	0.0005	0.0058	0.1212	0.6597
319	6	0.896	9.28	44.99	0.1001	0.6337	0.000	0.14	0.003	0.1250	0.6307	0.0585	0.0011	0.0074	0.6337	0.0548	0.0010	0.0075	0.1250	0.6157
	6	0.0628	0.0014	0.0076	0.1150	0.6073	0.0616	0.0010	0.0075	0.1250	0.6157	0.0628	0.0014	0.0076	0.6073	0.0616	0.0010	0.0075	0.1250	0.6157
319	7	0.899	12.40	44.99	0.0835	0.6081	0.002	0.15	0.004	0.1004	0.6176	0.0730	0.0012	0.0088	0.6081	0.0705	0.0010	0.0089	0.1004	0.6176
	6	0.0757	0.0013	0.0087	0.0920	0.5805	0.0767	0.0010	0.0089	0.1004	0.6176	0.0757	0.0013	0.0087	0.5805	0.0767	0.0010	0.0089	0.1004	0.6176
319	8	0.899	-0.000	44.99	-0.4176	-0.003	-0.006	0.06	-0.000	-0.3434	1.1015	0.0015	-0.0001	0.0003	-0.7746	-0.0016	-0.0001	-0.000	-0.8061	0.3939
	6	0.0012	-0.0003	0.0001	1.5097	0.5586	-0.0014	-0.0002	-0.000	-0.3434	1.1015	0.0012	-0.0003	0.0001	0.5586	-0.0014	-0.0002	-0.000	-0.8061	0.3939

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key →															
320	6	7	0.599	-5.09	-0.00	-0.0809	-0.02	-0.10	0.06	-0.07	0.1559	0.6900						
	8	7	0.0029	-0.0000	0.0002	-0.08920	0.4637	-0.0235	-0.0007	-0.0032	0.2289	0.6556						
	9	7	0.0005	0.0002	-0.0000	1.8920	-0.3379	-0.0227	-0.0010	-0.0029								
320	7	7	0.598	-0.03	-0.00	-0.0691	-0.03	-0.06	0.05	-0.03	-16.9477	-4.3110						
	8	7	0.0022	-0.0000	0.0004	-0.0691	0.9958	-0.0000	0.0002	0.0000	-0.3509	0.7309						
	9	7	0.0007	0.0001	-0.0000	1.3541	-0.3188	0.0017	-0.0001	0.0002								
320	8	7	0.599	1.01	-0.00	-0.0198	-0.03	-0.04	0.05	-0.02	0.2616	0.6194						
	8	7	0.0023	-0.0000	0.0004	-0.0198	0.7114	0.0082	0.0004	0.0010	0.0781	0.6688						
	9	7	0.0001	0.0001	0.0000	5.6087	1.5117	0.0083	0.0001	0.0011								
320	9	7	0.598	3.10	0.00	-0.0297	-0.02	-0.01	0.05	0.00	0.2621	0.7088						
	8	7	0.0030	-0.0000	0.0003	-0.0297	0.5404	0.0254	0.0013	0.0036	0.1318	0.6539						
	9	7	0.0001	0.0001	0.0000	4.9858	0.7438	0.0257	0.0006	0.0033								
320	10	7	0.598	6.16	-0.00	-0.0461	-0.02	0.01	0.06	0.03	0.2217	0.6474						
	8	7	0.0021	-0.0000	0.0002	-0.0461	0.5666	0.0508	0.0022	0.0065	0.1482	0.5794						
	9	7	0.0004	0.0002	0.0001	2.4372	1.7318	0.0538	0.0017	0.0067								
320	11	7	0.599	9.27	0.00	-0.0933	-0.03	0.05	0.05	0.05	0.1782	0.5896						
	8	7	0.0006	-0.0001	0.0003	-0.0933	2.6208	0.7333	0.0026	0.0086	0.1141	0.5631						
	9	7	0.0001	0.0001	0.0002	4.6607	5.1832	0.0735	0.0021	0.0085	0.0934	0.5610						
320	12	7	0.598	12.44	0.00	-0.1880	-0.03	0.05	0.05	0.06	0.1141	0.5631						
	8	7	0.0019	-0.0000	0.0001	-0.1880	0.4539	0.0846	0.0019	0.0095	0.1782	0.5896						
	9	7	0.0004	0.0002	0.0001	2.5128	1.3762	0.0856	0.0015	0.0096	0.1482	0.5794						
320	13	7	0.600	-0.03	-0.00	-0.0949	-0.02	-0.07	0.05	-0.03	0.6006	-0.2520						
	8	7	0.0021	-0.0000	0.0004	-0.0949	0.0060	0.0009	0.0001	-0.0000	-0.3727	0.6291						
	9	7	0.0000	0.0001	0.0000	20.6588	0.4668	0.0015	-0.0001	0.0002								
321	1	7	0.599	-3.09	44.99	0.2380	-0.05	-0.08	0.01	-0.05	0.1035	0.6449						
	6	7	-0.0153	-0.0007	-0.0018	0.2380	0.6042	-0.0170	-0.0003	-0.0021	0.2649	0.6849						
	9	7	-0.0152	-0.0004	-0.0017	0.1369	0.5772	-0.0144	-0.0007	-0.0019								
321	2	7	0.600	-0.00	44.99	-0.3022	-0.02	-0.07	0.05	-0.03	-0.5678	0.6201						
	6	7	0.0012	-0.0000	0.0003	-0.3022	1.4019	0.0015	0.0001	-0.0001	-0.5277	0.5020						
	9	7	0.0005	0.0002	0.0001	2.5345	1.0165	0.0012	-0.0001	0.0001								
321	3	7	0.600	1.03	44.99	0.1165	-0.02	-0.06	0.06	-0.02	1.3998	1.3040						
	6	7	0.0061	0.0001	0.0010	0.1165	0.8112	0.0016	0.0004	0.0004	0.0311	0.6292						
	9	7	0.0050	0.0004	0.0008	0.4109	0.8426	0.0062	0.0000	0.0007								

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key															
322	7	7	0.599	5.14	-0.0000	-0.0000	-0.0000	-0.0115	-0.02	6.08	0.04	0.0073	6.07	0.1968	0.6385			
	8	6	0.0024	-0.0000	0.0000	0.0000	0.0000	0.0000	0.7361	0.0575	0.0022	0.0073	0.1496	0.6202				
	9	9	0.0001	0.0000	0.0000	0.0000	0.0000	0.0000	3.7509	0.0588	0.0017	0.0072						
322	8	7	0.599	6.21	-0.0000	0.0000	-0.0889	-0.02	6.10	0.04	6.09	0.5836						
	8	6	0.0021	-0.0000	0.0000	0.0000	0.0000	0.0000	0.5333	0.0766	0.0025	0.0089	0.1653	0.5836				
	9	9	0.0002	0.0000	0.0000	0.0000	0.0000	0.0000	2.4657	0.0772	0.0020	0.0088	0.1321	0.5753				
322	9	7	0.599	9.29	-0.0000	0.0000	-0.3589	-0.02	6.10	0.04	6.09	0.5641						
	8	6	0.0013	-0.0000	0.0000	0.0000	0.0000	0.0000	0.392	0.0860	0.0017	0.0097	0.1003	0.5641				
	9	9	0.0006	0.0000	0.0000	0.0000	0.0000	0.0000	1.5770	0.0864	0.0013	0.0096	0.0797	0.5598				
322	10	7	0.599	12.44	-0.0000	0.0000	-0.3624	-0.02	6.08	0.04	6.06	0.5586						
	8	6	0.0013	-0.0000	0.0000	0.0000	0.0000	0.0000	1.1714	0.0823	0.0010	0.0091	0.0607	0.5586				
	9	9	0.0003	0.0000	0.0000	0.0000	0.0000	0.0000	3.2325	0.0816	0.0006	0.0090	0.0374	0.5559				
322	11	7	0.599	0.02	-0.0000	0.0000	-0.0620	-0.02	6.04	0.04	6.03	0.6744						
	8	6	0.0032	-0.0000	0.0000	0.0000	0.0000	0.0000	0.5752	0.0329	0.0013	0.0044	0.2009	0.6744				
	9	9	0.0003	0.0000	0.0000	0.0000	0.0000	0.0000	0.3580	0.0330	0.0008	0.0044	0.1270	0.6303				
323	1	7	0.599	-3.02	0.0003	0.0018	0.1440	0.03	6.03	6.01	6.06	0.7234						
	8	6	0.0135	0.0003	0.0018	0.0017	0.2871	0.6765	0.0119	0.0141	0.0002	0.0017	0.2728	0.6871				
	9	9	0.0118	0.0006	0.0017	0.0017	0.0000	0.7297	0.0141	0.0141	0.0002	0.0019	0.0952	0.6871				
323	2	7	0.599	0.07	0.0045	0.0042	0.1527	0.06	6.03	6.03	6.09	0.6779						
	8	6	0.0343	0.0010	0.0045	0.0042	0.2056	0.6332	0.0305	0.0333	0.0012	0.0041	0.2032	0.6779				
	9	9	0.0319	0.0013	0.0042	0.0042	0.0000	0.6718	0.0333	0.0333	0.0008	0.0044	0.1257	0.6733				
323	3	7	0.598	1.11	0.0014	0.0052	0.1549	0.07	6.05	6.05	6.11	0.6915						
	8	6	0.0404	0.0012	0.0054	0.0052	0.1933	0.6776	0.0363	0.0402	0.0015	0.0050	0.2136	0.6915				
	9	9	0.0385	0.0014	0.0052	0.0052	0.0000	0.6850	0.0402	0.0402	0.0010	0.0053	0.1308	0.6666				
323	4	7	0.599	3.17	0.0019	0.0068	0.1659	0.09	6.06	6.06	6.14	0.6340						
	8	6	0.0527	0.0017	0.0068	0.0065	0.1985	0.6451	0.0491	0.0516	0.0017	0.0062	0.1736	0.6340				
	9	9	0.0500	0.0019	0.0065	0.0065	0.0000	0.6496	0.0516	0.0516	0.0015	0.0066	0.1481	0.6460				
323	5	7	0.598	6.22	0.0025	0.0085	0.1562	0.11	6.09	6.09	6.16	0.607						
	8	6	0.0679	0.0021	0.0085	0.0085	0.1874	0.6277	0.0643	0.0662	0.0022	0.0080	0.1714	0.6290				
	9	9	0.0675	0.0025	0.0085	0.0085	0.0000	0.6019	0.0662	0.0662	0.0020	0.0079	0.1512	0.6025				
323	6	7	0.599	9.35	0.0023	0.0087	0.1435	0.14	6.10	6.10	6.17	0.608						
	8	6	0.0780	0.0022	0.0090	0.0090	0.1522	0.5823	0.0787	0.0775	0.0024	0.0092	0.1555	0.5875				
	9	9	0.0759	0.0023	0.0087	0.0087	0.0000	0.5769	0.0775	0.0775	0.0020	0.0088	0.1308	0.5688				

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key															
323	1	6	9	0.599	12.46	44.99	0.1097	0.5732	6.14	6.11	6.16	6.08	0.1323	0.5687				
				0.0851	0.0018	0.0097	0.1039	0.5700	0.0856	0.0022	0.0093	0.0097	0.0928	0.5613				
				0.0809	0.0016	0.0092	0.1039	0.5700	0.0836	0.0015	0.0015	0.0093	0.0928	0.5613				
323	8	6	9	0.599	0.06	44.99	0.1497	0.6762	6.06	6.03	6.10	6.02	0.1627	0.6290				
				0.0338	0.0010	0.0045	0.1946	0.6081	0.0321	0.0010	0.0008	0.0040	0.1250	0.6687				
				0.0313	0.0012	0.0043	0.1946	0.6081	0.0332	0.0008	0.0008	0.0044	0.1250	0.6687				
324	1	7	9	0.599	-2.94	44.99	0.1590	0.6376	15.11	15.09	15.08	15.07	0.1656	0.6300				
				0.0608	0.0019	0.0077	0.1544	0.6500	0.0608	0.0020	0.0015	0.0076	0.1236	0.6133				
				0.0583	0.0018	0.0075	0.1544	0.6500	0.0621	0.0015	0.0015	0.0076	0.1236	0.6133				
324	2	7	9	0.600	0.16	44.99	0.1466	0.5943	15.13	15.11	15.11	15.08	0.1496	0.6008				
				0.0755	0.0022	0.0089	0.1664	0.6017	0.0758	0.0022	0.0019	0.0091	0.1310	0.5875				
				0.0733	0.0024	0.0088	0.1664	0.6017	0.0756	0.0019	0.0019	0.0088	0.1310	0.5875				
324	3	7	9	0.599	1.18	44.99	0.1362	0.5871	15.14	15.12	15.11	15.09	0.1482	0.5898				
				0.0779	0.0021	0.0091	0.1596	0.5879	0.0788	0.0023	0.0019	0.0093	0.1260	0.5747				
				0.0755	0.0024	0.0088	0.1596	0.5879	0.0782	0.0019	0.0019	0.0089	0.1260	0.5747				
324	4	7	9	0.599	3.21	44.99	0.1085	0.5672	15.14	15.12	15.11	15.09	0.1314	0.5770				
				0.0830	0.0018	0.0094	0.1286	0.5743	0.0827	0.0021	0.0017	0.0095	0.1069	0.5645				
				0.0799	0.0020	0.0091	0.1286	0.5743	0.0826	0.0017	0.0017	0.0093	0.1069	0.5645				
324	5	7	9	0.599	5.29	44.99	0.0694	0.5647	15.14	15.11	15.10	15.08	0.0853	0.5712				
				0.0883	0.0012	0.0099	0.0872	0.5719	0.0890	0.0015	0.0012	0.0101	0.0615	0.5628				
				0.0831	0.0014	0.0095	0.0872	0.5719	0.0882	0.0012	0.0012	0.0099	0.0507	0.5528				
324	6	7	9	0.599	9.37	45.00	0.0475	0.5644	15.12	15.11	15.09	15.07	0.0451	0.5593				
				0.0846	0.0008	0.0095	0.0683	0.5658	0.0898	0.0011	0.0008	0.0101	0.0299	0.5569				
				0.0791	0.0010	0.0089	0.0683	0.5658	0.0821	0.0008	0.0005	0.0090	0.0299	0.5569				
324	7	7	9	0.599	12.45	45.00	0.1430	0.5885	15.12	15.10	15.08	15.07	0.1458	0.5958				
				0.0849	0.0005	0.0096	0.1696	0.6075	0.0878	0.0007	0.0005	0.0098	0.1314	0.5883				
				0.0824	0.0008	0.0094	0.1696	0.6075	0.0836	0.0005	0.0005	0.0093	0.1314	0.5883				
324	8	7	9	0.599	0.14	44.99	-0.1577	0.5738	15.14	15.11	15.11	15.09	0.1767	0.6471				
				0.0759	0.0021	0.0089	-1.5390	0.0151	0.0763	0.0022	0.0019	0.0090	0.1296	0.6294				
				0.0729	0.0024	0.0088	-1.5390	0.0151	0.0756	0.0019	0.0019	0.0088	0.1296	0.6294				
325	1	7	9	0.599	-2.98	-0.00	0.1577	0.5738	15.08	0.05	0.05	15.07	0.1767	0.6471				
				0.0025	-0.0000	0.0002	-0.1577	0.5738	0.0541	0.0019	0.0019	0.0070	0.1296	0.6294				
				-0.0005	-0.0001	-0.0000	-0.1577	0.0151	0.0567	0.0014	0.0014	0.0071	0.1296	0.6294				

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																
325	2	0.599	0.0031	-0.0002	0.0600	-0.0000	0.0004	-0.0000	-0.0650	-0.1319	-0.002	0.0766	0.0019	0.05	15.09	0.0091	0.1600	0.6024
	8	0.600	0.0029	-0.0000	1.09	-0.0004	0.0004	-0.0606	0.8179	-0.003	0.0789	0.0019	0.05	15.09	0.0092	0.1493	0.5834	
	9	0.599	0.0032	-0.0004	3.17	-0.0003	0.0000	-0.0607	4.1888	-0.002	0.0859	0.0014	0.05	15.09	0.0097	0.1206	0.5836	
325	4	0.599	0.0032	-0.0004	3.17	-0.0003	0.0000	-0.0607	4.1888	-0.002	0.0859	0.0014	0.05	15.09	0.0097	0.0986	0.5655	
	8	0.599	0.0021	-0.0000	6.22	-0.0002	0.0000	-0.1524	0.2753	-0.003	0.0844	0.0007	0.05	15.07	0.0093	0.0830	0.5576	
	9	0.599	0.0010	-0.0002	9.29	-0.0002	0.0000	-0.3805	2.7683	-0.002	0.0862	0.0003	0.05	15.06	0.0095	0.0657	0.5611	
325	6	0.600	0.0020	-0.0008	12.47	-0.0001	0.0000	-0.2042	0.5184	-0.003	0.0909	0.0002	0.05	15.06	0.0102	0.0401	0.5605	
	8	0.601	0.0033	-0.0001	0.05	-0.0003	0.0000	-0.0456	0.7259	-0.002	0.0917	0.0019	0.05	15.09	0.0101	0.0213	0.5524	
	9	1.248	0.0010	-0.0010	3.14	-0.0001	0.0001	-0.7546	1.7830	-0.002	0.0754	0.0019	0.06	3.00	0.0057	0.0350	0.5651	
327	3	1.249	0.0019	-0.0011	0.03	-0.0002	0.0001	-0.4333	0.6135	-0.003	0.0423	0.0001	0.06	2.99	0.0021	0.1620	0.6042	
	8	1.249	0.0016	-0.0010	0.99	-0.0002	0.0001	-0.4635	0.8809	-0.002	0.0140	0.0001	0.06	2.98	0.0009	0.1313	0.5842	
	9	1.249	0.0010	-0.0003	3.10	-0.0001	0.0003	-1.6996	0.8397	-0.002	0.0149	0.0001	0.06	2.96	0.0009	0.0487	0.6801	
327	4	1.249	0.0011	-0.0003	3.10	-0.0001	0.0003	-1.6996	0.8397	-0.002	0.0149	0.0001	0.06	2.96	0.0009	-0.0196	0.6947	
	8	1.249	0.0011	-0.0003	3.10	-0.0001	0.0003	-1.6996	0.8397	-0.002	0.0149	0.0001	0.06	2.96	0.0009	-0.0701	0.7124	
	9	1.249	0.0011	-0.0003	3.10	-0.0001	0.0003	-1.6996	0.8397	-0.002	0.0149	0.0001	0.06	2.96	0.0009	-0.0409	0.7215	
327	5	1.249	0.0016	-0.0010	0.99	-0.0002	0.0001	-0.4635	0.8099	-0.002	0.0050	0.0001	0.06	2.98	0.0007	-0.2092	0.7389	
	8	1.249	0.0011	-0.0001	3.10	-0.0001	0.0001	-0.5531	0.8479	-0.001	0.0114	0.0002	0.06	2.96	0.0015	-0.1510	0.8058	
	9	1.249	0.0011	-0.0001	3.10	-0.0001	0.0001	-0.5531	0.8479	-0.001	0.0114	0.0002	0.06	2.96	0.0011	0.1094	0.6670	

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key															
327	7	1.249	-0.0001	-2.4570	-0.02	-0.368	0.07	-2.95	0.0024	0.6734	0.0003	1.3673	0.0368	0.0000	0.0048	0.0048	0.0048	0.6539
	8	0.0003	0.0004	1.3149	0.9816	0.0372	-0.0003	0.0000	-0.0483	0.6539								
	9	0.0015	0.0004															
327	8	1.249	-0.0002	-1.8384	-0.03	-2.86	0.07	-2.93	0.0283	0.6600	0.0000	0.5116	0.0632	0.0003	0.0082	0.0082	0.0082	0.6510
	8	0.0005	0.0003	1.3196	0.9799	0.0634	-0.0006	0.0000	-0.0521	0.6510								
	9	0.0014	0.0003															
327	9	1.248	-0.0001	-3.1707	-0.03	-2.85	0.07	-2.92	0.0405	0.6387	0.0001	3.2457	0.0875	0.0007	0.0111	0.0111	0.0111	0.6387
	8	0.0002	0.0004	4.2661	1.7141	0.0874	-0.0010	0.0000	-0.0607	0.6387								
	9	0.0005	0.0004															
327	10	1.249	-0.0001	-0.4590	-0.02	-2.95	0.06	-2.98	0.0486	0.7479	0.0001	0.7081	0.0140	0.0001	0.0020	0.0022	0.0022	0.7479
	8	0.0018	0.0004	2.5000	1.2936	-0.0150	-0.0001	0.0000	-0.0521	0.7479								
	9	0.0008	0.0004															
328	1	1.249	-0.0001	-0.6735	-0.03	-0.09	0.05	-0.05	0.0321	0.6964	0.0001	0.6604	0.0264	0.0001	0.0036	0.0036	0.0036	0.6964
	8	0.0011	0.0003	3.4864	1.7016	-0.0258	-0.0000	0.0000	-0.0141	0.6964								
	9	0.0005	0.0003															
328	2	1.251	-0.0001	-0.4133	-0.02	-0.06	0.06	-0.03	0.0323	0.8335	0.0001	0.5496	0.0000	0.0002	0.0000	0.0000	0.0000	0.8335
	8	0.0020	0.0003	2.1112	1.0620	0.0000	-0.0002	0.0000	-0.0371	0.8335								
	9	0.0009	0.0003															
328	3	1.251	-0.0001	-0.4385	-0.02	-0.05	0.05	-0.02	0.0106	0.6563	0.0001	0.589	0.0091	0.0001	0.0009	0.0009	0.0009	0.6563
	8	0.0018	0.0003	2.2211	1.2025	0.0080	-0.0001	0.0000	-0.1003	0.6563								
	9	0.0008	0.0003															
328	4	1.251	-0.0001	-0.6018	-0.02	-0.03	0.05	-0.01	0.0129	0.6685	0.0001	0.615	0.0267	0.0000	0.0035	0.0035	0.0035	0.6685
	8	0.0011	0.0003	1.8623	1.1079	0.0252	-0.0002	0.0000	-0.0560	0.6685								
	9	0.0010	0.0003															
328	5	1.251	-0.0001	-2.9671	-0.02	-0.00	0.06	-0.00	0.0284	0.6640	0.0001	1.3995	0.0528	0.0003	0.0070	0.0070	0.0070	0.6640
	8	0.0003	0.0004	1.3774	0.8972	0.0531	-0.0005	0.0000	-0.0548	0.6640								
	9	0.0015	0.0004															
328	6	1.250	-0.0001	-3.5089	-0.02	0.00	0.07	-0.00	0.0358	0.6550	0.0001	1.5354	0.0774	0.0005	0.0101	0.0101	0.0101	0.6550
	8	0.0002	0.0001	1.8511	1.5768	0.0780	-0.0008	0.0000	-0.0568	0.6550								
	9	0.0010	0.0003															
328	7	1.249	-0.0001	-4.1172	-0.03	0.02	0.06	-0.00	0.0360	0.6249	0.0001	0.7781	0.0984	0.0007	0.0124	0.0124	0.0124	0.6249
	8	0.0008	0.0004	1.0755	1.7254	0.0993	-0.0012	0.0000	-0.0619	0.6249								
	9	0.0005	0.0004															

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																						
328	8	7	1.249	-0.0018	-0.0007	-0.0004	-0.0003	-0.0002	-0.0001	-0.4937	2.8230	-0.02	0.6435	1.2156	-0.0001	0.0001	-0.07	0.0001	0.06	-0.0001	-0.0001	4.0712	26.0916	-5.3723	18.8416
329	1	8	1.250	-0.0009	0.0003	-0.0004	-0.0004	-0.0001	0.0001	-0.8559	5.3635	-0.02	0.7257	1.7678	-0.0001	0.0001	0.98	0.0001	0.05	-0.0001	-0.0001	-0.0284	0.0385	0.7004	0.7221
329	2	9	1.249	0.0021	0.0007	0.0004	0.0000	0.0001	0.0002	-0.4508	2.7665	-0.02	0.5057	1.0310	0.0001	0.0002	1.01	0.0001	0.05	0.0001	0.0004	0.1705	-0.2410	0.6807	0.5069
329	3	7	1.249	0.0022	0.0010	0.0004	0.0001	0.0001	0.0001	-0.3519	2.7764	-0.02	0.3970	0.7292	0.0001	0.0002	1.02	0.0001	0.05	0.0001	0.0016	0.0402	-0.0775	0.6577	0.6286
329	4	6	1.249	0.0011	0.0010	0.0003	0.0001	0.0001	0.0002	-0.5528	1.8520	-0.03	0.6961	1.0209	0.0001	0.0003	1.04	0.0001	0.05	0.0001	0.0040	-0.0015	-0.0560	0.6721	0.6615
329	5	7	1.249	0.0004	0.0015	0.0004	0.0002	0.0001	0.0001	-2.2921	1.4221	-0.02	1.1309	0.7820	0.0001	0.0006	1.06	0.0001	0.06	0.0004	0.0076	-0.0357	-0.0572	0.6589	0.6548
329	6	6	1.247	0.0002	0.0014	0.0003	0.0002	0.0001	0.0002	-3.8500	1.3634	-0.03	0.8482	0.9265	0.0001	0.0006	1.08	0.0001	0.06	0.0006	0.0107	-0.0384	-0.0580	0.6490	0.6434
329	7	8	1.249	0.0006	0.0005	0.0004	0.0001	0.0001	0.0001	-1.3321	3.8238	-0.02	0.8013	1.5402	0.0001	0.0012	1.11	0.0001	0.06	0.0007	0.0128	-0.0365	-0.0622	0.6263	0.6219
329	8	9	1.249	0.0019	0.0006	0.0004	0.0001	0.0001	0.0002	-0.5067	3.4716	-0.02	0.6517	1.6095	0.0001	0.0002	1.01	0.0001	0.05	0.0001	0.0004	0.1213	-0.2525	0.5949	0.4982
330	1	7	1.248	0.0012	0.0004	0.0004	0.0001	0.0001	0.0001	-0.6530	4.4752	-0.02	0.5699	1.3209	0.0001	0.0002	2.96	0.0001	0.05	0.0001	3.03	-0.0127	0.0147	0.7125	0.7448
330	2	6	1.248	0.0021	0.0008	0.0004	0.0001	0.0001	0.0001	-0.4395	2.5510	-0.02	0.4607	0.9758	0.0001	0.0002	2.99	0.0001	0.05	0.0001	3.05	0.0244	-0.0907	0.6711	0.6484

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																				
330	3	7	1.248	1.05	-0.002	-0.5088	-0.02	3.00	0.05	3.05	0.0056	0.6746	0.0017	-0.0001	0.0004	0.0002	-0.7521	0.0255	0.0000	0.0034	0.0051	0.6638
330	4	7	1.249	5.17	-0.001	-0.4077	-0.02	3.03	0.05	3.06	-0.0201	0.6720	0.0008	0.0004	0.0004	0.0001	0.3308	0.0440	-0.0001	0.0059	-0.0619	0.6625
330	5	7	1.249	6.30	-0.001	-0.1026	-0.02	3.05	0.06	3.09	-0.0398	0.6580	0.0012	0.0004	0.0004	0.0001	4.1548	0.0691	-0.0008	0.0090	-0.0583	0.6540
330	6	7	1.249	9.43	-0.002	-4.3253	-0.02	3.07	0.06	3.09	-0.0394	0.6406	0.0012	0.0004	0.0004	0.0002	1.1347	0.0922	-0.0011	0.0117	-0.0609	0.6353
330	7	7	1.248	12.63	-0.001	-0.9802	-0.02	3.09	0.06	3.11	-0.0411	0.5252	0.0005	0.0004	0.0004	0.0001	1.4042	0.1110	-0.0014	0.0137	-0.0636	0.6181
330	8	7	1.250	-0.01	-0.002	-0.4728	-0.02	2.99	0.05	3.05	-0.0283	0.6661	0.0006	0.0004	0.0004	0.0001	0.6693	0.0151	-0.0002	0.0020	-0.0929	0.6424
331	1	7	1.249	-3.08	-0.001	-0.8215	-0.02	6.00	0.05	5.99	-0.0482	0.6949	0.0004	0.0004	0.0004	0.0001	0.7619	0.0048	0.0000	0.0006	-0.0455	0.6046
331	2	7	1.248	0.02	-0.002	-0.4801	-0.02	6.03	0.05	6.01	-0.0133	0.6860	0.0007	0.0004	0.0004	0.0001	1.0938	0.0314	-0.0004	0.0042	-0.0749	0.6740
331	3	7	1.248	1.05	-0.002	-0.4137	-0.02	6.04	0.05	6.01	-0.0249	0.6759	0.0008	0.0004	0.0004	0.0001	0.5001	0.421	-0.0002	0.0054	-0.0672	0.6770
331	4	7	1.250	3.18	-0.001	-0.7475	-0.02	6.06	0.05	6.01	-0.0389	0.6649	0.0009	0.0004	0.0004	0.0001	0.9196	0.0576	-0.0007	0.0077	-0.0620	0.6690
331	5	7	1.250	6.31	-0.002	-2.9690	-0.02	6.08	0.06	6.03	-0.0426	0.6504	0.0015	0.0004	0.0004	0.0001	1.1322	0.0833	-0.0010	0.0107	-0.0609	0.6544

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																			
331	6	7	1.250	9.44	-0.000	-5.9324	-0.02	6.10	0.06	6.05	-0.0385	0.6299	0.0001	-0.0001	0.0000	0.0000	0.0000	0.1032	-0.0007	0.0130	-0.0605	0.6230
		6	0.0012	0.0004	0.0002	1.7984	1.0609	0.1039	0.0012	0.0129	-0.0605	0.6230	0.0001	-0.0001	0.0000	0.0000	0.0000	0.1039	-0.0012	0.0129	-0.0605	0.6230
331	7	7	1.249	12.67	-0.001	-1.2666	-0.02	6.12	0.05	6.06	-0.0466	0.6170	0.0006	-0.0004	0.0001	0.0001	0.0001	0.1208	-0.0015	0.0147	-0.0660	0.6117
		6	0.0005	0.0004	0.0001	4.5110	1.4058	0.1208	-0.0015	0.0147	-0.0660	0.6117	0.0005	-0.0004	0.0001	0.0001	0.0001	0.1208	-0.0015	0.0147	-0.0660	0.6117
331	8	7	1.248	-0.00	-0.002	-0.4935	-0.02	6.03	0.05	6.00	-0.0207	0.6728	0.0019	-0.0004	0.0001	0.0001	0.0001	0.0320	-0.0004	0.0043	-0.0748	0.6743
		6	0.0005	0.0004	0.0001	3.9301	1.2713	0.0320	-0.0004	0.0043	-0.0748	0.6728	0.0019	-0.0004	0.0001	0.0001	0.0001	0.0320	-0.0004	0.0043	-0.0748	0.6743
332	1	7	1.249	-3.05	-0.001	-0.6636	-0.02	9.06	0.05	8.98	-0.0167	0.6978	0.0012	-0.0004	0.0001	0.0001	0.0001	0.0205	-0.0003	0.0027	-0.0897	0.6941
		6	0.0004	0.0004	0.0001	4.7458	1.2065	0.0205	-0.0003	0.0027	-0.0897	0.6941	0.0012	-0.0004	0.0001	0.0001	0.0001	0.0205	-0.0003	0.0027	-0.0897	0.6941
332	2	7	1.248	0.05	-0.002	-0.4735	-0.02	9.09	0.05	8.99	-0.0382	0.6813	0.0019	-0.0004	0.0001	0.0001	0.0001	0.0489	-0.0003	0.0066	-0.0732	0.6781
		6	0.0009	0.0004	0.0001	2.3584	0.6756	0.0475	-0.0006	0.0064	-0.0732	0.6813	0.0019	-0.0004	0.0001	0.0001	0.0001	0.0489	-0.0003	0.0066	-0.0732	0.6781
332	3	7	1.248	1.09	-0.001	-0.4217	-0.02	9.10	0.05	9.01	-0.0399	0.6719	0.0019	-0.0004	0.0001	0.0001	0.0001	0.0565	-0.0007	0.0076	-0.0680	0.6753
		6	0.0008	0.0004	0.0001	2.7573	0.8741	0.0565	-0.0007	0.0076	-0.0680	0.6753	0.0019	-0.0004	0.0001	0.0001	0.0001	0.0565	-0.0007	0.0076	-0.0680	0.6753
332	4	7	1.248	5.21	-0.001	-0.5696	-0.02	9.11	0.05	9.02	-0.0402	0.6630	0.0012	-0.0004	0.0001	0.0001	0.0001	0.0750	-0.0009	0.0096	-0.0633	0.6619
		6	0.0009	0.0004	0.0001	2.4158	1.0186	0.0725	-0.0009	0.0096	-0.0633	0.6619	0.0012	-0.0004	0.0001	0.0001	0.0001	0.0750	-0.0009	0.0096	-0.0633	0.6619
332	5	7	1.248	6.35	-0.001	-2.6418	-0.02	9.13	0.05	9.02	-0.0413	0.6422	0.0003	-0.0004	0.0001	0.0001	0.0001	0.0962	-0.0011	0.0122	-0.0616	0.6354
		6	0.0015	0.0004	0.0002	1.5613	0.8248	0.0962	-0.0011	0.0122	-0.0616	0.6354	0.0003	-0.0004	0.0001	0.0001	0.0001	0.0962	-0.0011	0.0122	-0.0616	0.6354
332	6	7	1.248	9.49	-0.001	-2.4485	-0.02	9.15	0.05	9.04	-0.0420	0.6264	0.0004	-0.0002	0.0001	0.0001	0.0001	0.1143	-0.0014	0.0141	-0.0611	0.6195
		6	0.0015	0.0004	0.0002	1.4464	0.7491	0.1145	-0.0014	0.0141	-0.0611	0.6195	0.0004	-0.0002	0.0001	0.0001	0.0001	0.1143	-0.0014	0.0141	-0.0611	0.6195
332	7	7	1.250	12.70	-0.001	-1.4133	-0.02	9.16	0.05	9.04	-0.0406	0.6101	0.0006	-0.0004	0.0001	0.0001	0.0001	0.1297	-0.0012	0.0158	-0.0651	0.6073
		6	0.0007	0.0004	0.0001	3.3449	0.8209	0.1289	-0.0016	0.0158	-0.0651	0.6073	0.0006	-0.0004	0.0001	0.0001	0.0001	0.1289	-0.0016	0.0158	-0.0651	0.6073
332	8	7	1.251	0.02	-0.002	-0.5266	-0.02	9.08	0.05	9.00	-0.0387	0.6739	0.0019	-0.0002	0.0001	0.0001	0.0001	0.0486	-0.0003	0.0064	-0.0727	0.6809
		6	0.0009	0.0004	0.0001	2.7498	0.8575	0.0471	-0.0006	0.0064	-0.0727	0.6809	0.0019	-0.0002	0.0001	0.0001	0.0001	0.0486	-0.0003	0.0064	-0.0727	0.6809

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Pt.	Cd	See Key																		
333	1	7	1.250	-5.00	-0.00	-0.6094	-0.02	15.04	0.05	15.02	-0.0469	0.6894	0.0014	-0.0001	0.4205	0.0520	-0.0004	0.0071	-0.0071	-0.0763	0.6871
	6	9	0.0004	0.0004	0.0000	5.4891	1.1486	0.0522	-0.0007	0.0071	-0.0763	0.6871	0.0004	0.0000	0.4205	0.0522	-0.0007	0.0071	-0.0071	-0.0763	0.6871
	9	7	1.250	0.07	-0.00	-0.4748	-0.02	15.08	0.05	15.04	-0.0464	0.6655	0.0020	0.0001	0.4825	0.0784	-0.0007	0.0104	-0.0102	-0.0706	0.6598
333	2	7	1.250	0.07	-0.00	-0.4748	-0.02	15.08	0.05	15.04	-0.0464	0.6655	0.0008	0.0004	0.7369	0.0775	-0.0010	0.0102	-0.0102	-0.0706	0.6598
	6	9	0.0006	0.0004	0.0001	2.8721	0.7369	0.0775	-0.0010	0.0102	-0.0706	0.6598	0.0006	0.0004	0.7369	0.0775	-0.0010	0.0102	-0.0102	-0.0706	0.6598
	9	7	1.249	1.11	-0.00	-0.4219	-0.02	15.09	0.05	15.04	-0.0414	0.6584	0.0019	0.0001	0.5327	0.0862	-0.0007	0.0113	-0.0111	-0.0684	0.6512
333	3	7	1.249	1.11	-0.00	-0.4219	-0.02	15.09	0.05	15.04	-0.0414	0.6584	0.0006	0.0004	1.2015	0.0852	-0.0011	0.0111	-0.0111	-0.0684	0.6512
	6	9	0.0006	0.0004	0.0001	3.8652	1.2015	0.0852	-0.0011	0.0111	-0.0684	0.6512	0.0006	0.0004	1.2015	0.0852	-0.0011	0.0111	-0.0111	-0.0684	0.6512
	9	7	1.249	3.24	-0.00	-0.6808	-0.02	15.10	0.05	15.05	-0.0421	0.6395	0.0009	0.0004	0.8371	0.1006	-0.0012	0.0125	-0.0125	-0.0636	0.6349
333	4	7	1.249	3.24	-0.00	-0.6808	-0.02	15.10	0.05	15.05	-0.0421	0.6395	0.0009	0.0004	0.8371	0.1006	-0.0012	0.0125	-0.0125	-0.0636	0.6349
	6	9	0.0009	0.0004	0.0001	2.3732	0.8371	0.1006	-0.0012	0.0125	-0.0636	0.6349	0.0009	0.0004	0.8371	0.1006	-0.0012	0.0125	-0.0125	-0.0636	0.6349
	9	7	1.249	6.37	-0.00	-1.4271	-0.02	15.12	0.05	15.06	-0.0437	0.6257	0.0007	0.0002	0.2558	0.1172	-0.0010	0.0146	-0.0144	-0.0588	0.6154
333	5	7	1.249	6.37	-0.00	-1.4271	-0.02	15.12	0.05	15.06	-0.0437	0.6257	0.0007	0.0002	0.2558	0.1172	-0.0010	0.0146	-0.0144	-0.0588	0.6154
	6	9	0.0016	0.0005	0.0002	1.5998	0.6536	0.1170	-0.0013	0.0144	-0.0588	0.6154	0.0016	0.0005	0.6536	0.1170	-0.0013	0.0144	-0.0144	-0.0588	0.6154
	9	7	1.249	9.47	-0.00	-3.8812	-0.02	15.13	0.05	15.07	-0.0507	0.6125	0.0002	0.0002	0.9238	0.1313	-0.0016	0.0159	-0.0159	-0.0535	0.6058
333	6	7	1.249	9.47	-0.00	-3.8812	-0.02	15.13	0.05	15.07	-0.0507	0.6125	0.0002	0.0002	0.9238	0.1313	-0.0016	0.0159	-0.0159	-0.0535	0.6058
	6	9	0.0013	0.0004	0.0002	1.7838	0.9238	0.1313	-0.0016	0.0159	-0.0535	0.6058	0.0013	0.0004	0.9238	0.1313	-0.0016	0.0159	-0.0159	-0.0535	0.6058
	9	7	1.250	12.70	-0.00	-1.0131	-0.02	15.13	0.05	15.06	-0.0444	0.5981	0.0008	0.0001	0.6012	0.1352	-0.0012	0.0161	-0.0158	-0.0641	0.5939
333	7	7	1.250	12.70	-0.00	-1.0131	-0.02	15.13	0.05	15.06	-0.0444	0.5981	0.0008	0.0001	0.6012	0.1352	-0.0012	0.0161	-0.0158	-0.0641	0.5939
	6	9	0.0006	0.0004	0.0001	3.4619	0.9905	0.1337	-0.0017	0.0158	-0.0641	0.5939	0.0006	0.0001	0.9905	0.1337	-0.0017	0.0158	-0.0158	-0.0641	0.5939
	9	7	1.250	12.70	-0.00	-1.0131	-0.02	15.13	0.05	15.06	-0.0444	0.5981	0.0008	0.0001	0.6012	0.1352	-0.0012	0.0161	-0.0158	-0.0641	0.5939
334	3	7	0.898	-3.10	-0.00	-0.2157	-0.02	-3.00	0.06	-3.01	0.0737	0.6549	0.0023	0.0001	0.6980	-0.0483	-0.0007	-0.0063	0.0057	0.1364	0.6422
	6	9	0.0005	0.0002	-0.0000	2.3045	-0.4776	-0.0448	-0.0012	-0.0057	0.1364	0.6422	0.0005	0.0002	-0.4776	-0.0448	-0.0012	-0.0057	0.0057	0.1364	0.6422
	9	7	0.899	-0.06	-0.00	-0.0535	-0.02	-2.96	0.06	-2.98	0.0634	0.6775	0.0032	0.0000	0.6907	-0.0154	-0.0001	-0.0021	0.0021	0.1949	0.6775
334	4	7	0.899	-0.06	-0.00	-0.0535	-0.02	-2.96	0.06	-2.98	0.0634	0.6775	0.0032	0.0000	0.6907	-0.0154	-0.0001	-0.0021	0.0021	0.1949	0.6775
	6	9	0.0009	0.0002	0.0000	1.4013	0.2117	-0.0158	-0.0006	-0.0021	0.1949	0.6775	0.0009	0.0002	0.2117	-0.0158	-0.0006	-0.0021	0.0021	0.1949	0.6775
	9	7	0.900	0.99	-0.00	-0.0402	-0.01	-2.94	0.06	-2.97	-0.0207	0.7026	0.0031	0.0000	0.7255	-0.0057	0.0000	-0.0008	0.0008	-0.2798	0.7026
334	5	7	0.900	0.99	-0.00	-0.0402	-0.01	-2.94	0.06	-2.97	-0.0207	0.7026	0.0031	0.0000	0.7255	-0.0057	0.0000	-0.0008	0.0008	-0.2798	0.7026
	6	9	0.0005	0.0002	0.0001	2.3856	1.0742	-0.0062	-0.0003	-0.0008	-0.2798	0.6993	0.0005	0.0002	1.0742	-0.0062	-0.0003	-0.0008	0.0008	-0.2798	0.6993
	9	7	0.900	5.09	-0.00	-0.0127	-0.02	-2.91	0.06	-2.95	0.2611	0.6608	0.0025	0.0000	0.8541	0.0117	0.0006	0.0015	0.0014	0.0740	0.6608
334	6	7	0.900	5.09	-0.00	-0.0127	-0.02	-2.91	0.06	-2.95	0.2611	0.6608	0.0025	0.0000	0.8541	0.0117	0.0006	0.0015	0.0014	0.0740	0.6608
	6	9	0.0010	0.0002	0.0001	1.3877	0.4973	0.0106	0.0001	0.0014	0.0740	0.6608	0.0010	0.0002	0.4973	0.0106	0.0001	0.0014	0.0014	0.0740	0.6608
	9	7	0.900	5.09	-0.00	-0.0127	-0.02	-2.91	0.06	-2.95	0.2611	0.6608	0.0025	0.0000	0.8541	0.0117	0.0006	0.0015	0.0014	0.0740	0.6608

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																				
334	7	0.898	6.19	-0.00	-0.02	-2.87	0.06	-2.92	0.1371	0.6558	0.0021	-0.0002	-0.003	-0.002	0.6977	0.7215	0.0429	0.0011	0.0056	0.0651	0.6733	
	8	0.0015	0.0002	0.0002	0.9321	0.0460	0.0006	0.0062			0.0015	0.0002	0.0003	0.7215	0.0460	0.0011	0.0056	0.0006				
334	7	0.900	9.30	-0.00	-0.02	-2.84	0.06	-2.89	0.1086	0.6110	0.0016	-0.0003	-0.003	-0.002	0.9356	0.9927	0.0673	0.0014	0.0082	0.0689	0.6048	
	8	0.0017	0.0002	0.0003	0.8632	0.0697	0.0009	0.0084			0.0017	0.0003	0.0003	0.9927	0.0697	0.0014	0.0082	0.0009				
334	7	0.898	12.46	0.00	-0.02	-2.83	0.07	-2.89	0.0645	0.5741	0.0016	0.0001	0.002	-0.002	0.8503	0.8562	0.0859	0.0011	0.0098	0.0448	0.5786	
	8	0.0011	0.0003	0.0001	1.3593	0.0872	0.0007	0.0101			0.0011	0.0001	0.002	0.8562	0.0872	0.0011	0.0098	0.0007				
334	10	0.900	-0.04	-0.00	-0.02	-2.96	0.06	-2.98	0.0478	0.6803	0.0029	0.0004	0.004	-0.002	0.7614	1.3254	-0.0157	0.0001	-0.0021	0.1694	0.6871	
	9	0.0004	0.0002	0.0001	3.1593	-0.0159	-0.0006	-0.0021			0.0004	0.0001	0.004	1.3254	-0.0159	-0.0006	-0.0021	-0.0006				
335	1	0.899	-3.08	-0.00	-0.02	-0.10	0.06	-0.05	0.0672	0.6734	0.0027	0.0003	0.003	-0.002	0.6409	9.1485	-0.0308	0.0004	-0.0041	0.1342	0.6808	
	9	0.0000	0.0002	0.0000	97.1715	-0.0297	-0.0007	-0.0040			0.0027	0.0003	0.003	9.1485	-0.0297	-0.0007	-0.0040	-0.0007				
335	2	0.899	-0.03	-0.00	-0.02	-0.07	0.06	-0.03	-2.0466	0.2527	0.0030	0.0004	0.004	-0.002	0.7615	0.5353	-0.0007	0.0003	-0.0000	0.7607	0.2527	
	9	0.0010	0.0002	0.0001	1.1894	0.0012	-0.0001	0.0001			0.0010	0.0001	0.004	0.5353	0.0012	-0.0001	0.0003	-0.0001				
335	3	0.899	1.01	-0.00	-0.02	-0.04	0.06	-0.02	0.4312	0.7905	0.0030	0.0004	0.004	-0.002	0.7455	0.4725	0.0078	0.0006	0.0012	0.0900	0.7223	
	9	0.0008	0.0002	0.0000	1.6784	0.0083	0.0001	0.0001			0.0030	0.0004	0.004	0.4725	0.0083	0.0006	0.0012	0.0001				
335	4	0.900	3.11	-0.00	-0.02	-0.03	0.06	-0.01	0.1255	0.6671	0.0025	0.0004	0.004	-0.002	0.7788	0.3174	0.0325	0.0008	0.0042	0.0502	0.6810	
	9	0.0012	0.0002	0.0000	1.1904	0.0312	0.0003	0.0003			0.0025	0.0004	0.004	0.3174	0.0312	0.0008	0.0042	0.0003				
335	5	0.898	6.19	-0.00	-0.02	0.00	0.07	0.02	0.1332	0.6328	0.0020	0.0002	0.002	-0.002	0.6570	0.7760	0.0567	0.0015	0.0071	0.0812	0.6294	
	9	0.0015	0.0002	0.0002	0.9379	0.0590	0.0009	0.0074			0.0020	0.0002	0.002	0.7760	0.0590	0.0015	0.0071	0.0009				
335	6	0.896	9.26	-0.00	-0.02	0.01	0.07	0.04	0.0795	0.5875	0.0017	0.0002	0.002	-0.002	0.7848	0.8698	0.0796	0.0012	0.0093	0.0570	0.5837	
	9	0.0017	0.0003	0.0002	0.9523	0.0809	0.0009	0.0094			0.0017	0.0002	0.002	0.8698	0.0809	0.0012	0.0093	0.0009				
335	7	0.899	12.45	0.00	-0.02	0.01	0.07	0.03	0.0350	0.5797	0.0018	0.0001	0.002	-0.002	0.7078	0.5920	0.01	0.0006	0.0103	0.0350	0.5797	
	9	0.0013	0.0003	0.0001	1.0972	0.0907	0.0003	0.0104			0.0018	0.0001	0.002	0.5920	0.0907	0.0006	0.0103	0.0003				

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key													
335	7 6 9	0.897 0.0032 0.0006	-0.004 0.0000 0.0003	-0.004 0.0000 0.0003	-0.0264 2.5437	-0.01 0.6258 0.5394	-0.001 0.0014	-0.06 0.0001	0.06 0.0001	-0.002 0.0001	-0.003 0.0001	-0.4130 0.6141	1.9369 0.6952		
336	1 6 9	0.899 0.0025 0.0007	-3.07 -0.0000 0.0002	-0.003 0.0000 -0.0000	-0.1888 1.8388	-0.02 0.7458 -0.0277	0.96 -0.0222	0.06 -0.0007	0.06 -0.0007	0.93 -0.0030 -0.0029	0.0501 0.1641	0.6427 0.6643			
336	2 6 9	0.898 0.0031 0.0008	-0.003 0.0000 0.0002	-0.004 0.0000 0.0000	-0.0464 1.5557	-0.02 0.7251 0.4986	1.01 0.0040 0.0054	0.06 0.0005 0.0000	0.06 0.0005 0.0000	0.96 0.0006 0.0007	0.6884 0.0294	0.8551 0.6952			
336	3 6 9	0.898 0.0031 0.0009	1.03 -0.0000 0.0002	-0.004 0.0000 0.0000	-0.0112 1.2615	-0.02 0.6758 0.3743	1.02 0.0155 0.0157	0.06 0.0006 0.0001	0.06 0.0006 0.0001	0.97 0.0022 0.0022	0.2239 0.0540	0.7128 0.7009			
336	4 6 9	0.897 0.0027 0.0009	3.12 -0.0000 0.0002	-0.003 0.0000 0.0001	-0.0004 1.2688	-0.01 0.6858 0.5201	1.04 0.378 0.0381	0.06 0.0010 0.0004	0.06 0.0010 0.0004	0.99 0.0052 0.0052	0.1452 0.0549	0.6878 0.6853			
336	5 6 9	0.898 0.0016 0.0014	6.21 -0.0000 0.0003	-0.003 0.0002 0.0002	-0.2183 1.0427	-0.02 1.0138 0.7675	1.07 0.0625 0.0638	0.07 0.0014 0.0009	0.07 0.0014 0.0009	1.02 0.0077 0.0078	0.1185 0.0773	0.6208 0.6171			
336	6 6 9	0.898 0.0015 0.0016	9.25 -0.0001 0.0002	-0.002 0.0002 0.0002	-0.3949 0.9121	-0.02 0.9610 0.8372	1.08 0.0820 0.0832	0.07 0.0013 0.0009	0.07 0.0013 0.0009	1.02 0.0095 0.0095	0.0824 0.0554	0.5848 0.5757			
336	7 6 9	0.900 0.0017 0.0017	12.45 -0.0001 0.0003	0.002 0.0001 0.0001	-0.3358 1.0371	-0.03 0.8176 0.3600	1.08 0.0897 0.0909	0.07 0.0006 0.0002	0.07 0.0006 0.0002	1.02 0.0105 0.0104	0.0361 0.0118	0.5863 0.5735			
336	8 6 9	0.897 0.0029 0.0007	-0.003 -0.0000 0.0003	-0.004 0.0000 0.0000	-0.0708 2.1745	-0.02 0.7710 0.4952	1.00 0.0042 0.0053	0.06 0.0005 0.0000	0.06 0.0005 0.0000	0.96 0.0006 0.0007	0.6492 0.0356	0.8007 0.6848			
337	1 6 9	0.898 0.0025 0.0005	-3.07 -0.0000 0.0002	-0.003 0.0000 0.0000	-0.1883 2.3297	-0.02 0.7643 0.1608	2.95 -0.0091 -0.0083	0.06 -0.0004 -0.0004	0.06 -0.0004 -0.0004	3.03 -0.0012 -0.0011	0.0960 0.2939	0.6732 0.6844			
337	2 6 9	0.899 0.0030 0.0007	-0.002 -0.0000 0.0002	-0.004 0.0000 0.0000	-0.0476 1.5539	-0.02 0.6982 0.5009	2.99 0.0182 0.0195	0.06 0.0006 0.0001	0.06 0.0006 0.0001	3.07 0.0025 0.0026	0.1897 0.0388	0.7119 0.6852			

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																								
337	3	0.898	1.02	-0.004	-0.0004	-0.0080	-0.02	3.02	0.06	3.02	0.0008	0.0042	0.1357	0.6940	0.0029	-0.0000	-0.0001	0.0002	0.0000	0.0004	0.7262	0.0305	0.0008	0.0042	0.0507	0.6997
337	4	0.899	3.12	0.0003	0.0003	0.0076	-0.02	3.04	0.06	3.10	0.0022	0.0062	0.1376	0.6464	0.0009	0.0002	0.0001	0.0002	0.0000	0.0003	0.8690	0.0479	0.0013	0.0065	0.0695	0.6617
337	5	0.899	6.19	-0.0003	-0.0003	-0.1991	-0.02	3.06	0.07	3.12	0.0015	0.0067	0.0943	0.6037	0.0018	0.0003	0.0002	0.0003	0.0000	0.0003	1.0334	0.0723	0.0013	0.0086	0.0630	0.5994
337	6	0.899	9.30	0.0002	0.0002	-0.2548	0.02	3.07	0.07	3.13	0.0021	0.0098	0.0599	0.5862	0.0016	0.0003	0.0002	0.0003	0.0000	0.0002	0.8831	0.0840	0.0010	0.0098	0.0374	0.5769
337	7	0.899	12.44	0.0002	0.0002	-0.5168	-0.02	3.07	0.07	3.12	0.0010	0.0108	0.0265	0.5835	0.0008	0.0002	0.0002	0.0002	0.0000	0.0003	1.3807	0.0929	0.0004	0.0107	0.0067	0.5742
337	8	0.898	0.00	0.0004	0.0004	-0.0406	0.02	3.01	0.06	3.07	0.0030	0.0026	0.2016	0.7203	0.0009	0.0002	0.0000	0.0002	0.0000	0.0004	0.7145	0.0182	0.0007	0.0027	0.0493	0.6997
338	1	0.898	-3.04	-0.0002	-0.0002	-0.1767	-0.02	6.00	0.06	6.00	0.0025	0.0008	0.5791	0.6923	0.0005	0.0000	-0.0000	0.0000	0.0000	0.0003	0.7340	0.0045	0.0005	0.0010	0.0606	0.7657
338	2	0.897	0.02	0.0004	0.0004	-0.0316	0.01	6.04	0.06	6.03	0.0028	0.0052	0.1222	0.6881	0.0007	0.0002	0.0000	0.0002	0.0000	0.0004	0.7445	0.0384	0.0009	0.0054	0.0506	0.6925
338	3	0.901	1.03	0.0002	0.0002	0.0116	-0.02	6.06	0.06	6.04	0.0027	0.0060	0.1409	0.6615	0.0010	0.0002	0.0000	0.0002	0.0000	0.0004	0.7783	0.0458	0.0012	0.0063	0.0650	0.6680
338	4	0.897	3.11	0.0002	0.0002	0.0481	0.02	6.07	0.06	6.06	0.0029	0.0079	0.1164	0.6304	0.0009	0.0002	0.0001	0.0002	0.0000	0.0003	0.6159	0.0632	0.0014	0.0080	0.0702	0.6273
338	5	0.898	6.20	-0.0002	-0.0002	-0.2880	-0.02	6.08	0.06	6.07	0.0017	0.0093	0.0839	0.5867	0.0015	0.0002	0.0002	0.0002	0.0000	0.0002	0.8278	0.0798	0.0013	0.0094	0.0605	0.5852

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key															
338	6	7	0.898	9.31	-0.002	-0.2901	-0.02	6.08	0.07	6.06	0.0341	0.5790						
	8	9	0.0016	-0.0003	0.0002	0.8194	0.7568	0.0883	0.0006	0.0102	0.0165	0.5760						
			0.0018	0.0003	0.0002		0.7668	0.0893	0.0002	0.0102								
338	7	7	0.898	12.44	0.002	-0.3518	-0.02	6.08	0.07	6.07	0.0117	0.5773						
	8	9	0.0018	-0.0001	0.0001	1.1772	0.6468	0.0980	0.0002	0.0113	0.0019	0.5749						
			0.0013	0.0003	0.0001		0.7241	0.0966	0.0000	0.0111								
338	8	7	0.898	0.01	-0.004	-0.0393	-0.02	6.04	0.06	6.04	0.1257	0.6957						
	8	9	0.0029	-0.0002	0.0000	1.3339	0.7514	0.0382	0.0009	0.0053	0.0521	0.7012						
			0.0008	0.0002	0.0000		0.3823	0.0387	0.0004	0.0054								
339	1	7	0.899	-3.02	-0.003	-0.1762	-0.02	9.07	0.06	9.00	0.1464	0.7377						
	8	9	0.0025	-0.0002	0.0000	1.8064	0.7529	0.0252	0.0007	0.0037	0.0369	0.7251						
			0.0006	0.0002	-0.0000		0.0137	0.0274	0.0002	0.0039								
339	2	7	0.899	0.02	-0.003	-0.0348	-0.02	9.10	0.06	9.02	0.1203	0.6479						
	8	9	0.0028	-0.0002	0.0000	1.3212	0.6909	0.0531	0.0012	0.0068	0.0676	0.6533						
			0.0009	0.0002	0.0000		0.3092	0.0539	0.0007	0.0070								
339	3	7	0.899	1.05	-0.003	0.0378	-0.02	9.11	0.06	9.04	0.1115	0.6330						
	8	9	0.0028	0.0002	0.0000	1.4557	0.6785	0.0617	0.0013	0.0078	0.0672	0.6383						
			0.0007	0.0002	0.0000		0.6290	0.0617	0.0008	0.0078								
339	4	7	0.900	3.14	-0.003	0.0294	-0.02	9.13	0.06	9.05	0.0898	0.6085						
	8	9	0.0022	0.0002	0.0001	1.3894	0.7719	0.0756	0.0013	0.0092	0.0552	0.6034						
			0.0008	0.0002	0.0001		0.7154	0.0754	0.0008	0.0091								
339	5	7	0.898	6.23	-0.003	-0.1898	-0.02	9.12	0.06	9.05	0.0432	0.5792						
	8	9	0.0014	-0.0002	0.0002	0.7312	1.0916	0.0849	0.0007	0.0098	0.0305	0.5795						
			0.0018	0.0002	0.0002		0.7419	0.0871	0.0005	0.0101								
339	6	7	0.897	9.33	-0.003	-0.4252	-0.02	9.13	0.07	9.04	0.0244	0.5798						
	8	9	0.0011	-0.0001	0.0003	0.6576	1.3685	0.0932	0.0004	0.0108	0.0060	0.5759						
			0.0021	0.0002	0.0003		0.7793	0.0938	0.0001	0.0108								
339	7	7	0.899	12.46	0.002	-0.3638	-0.02	9.13	0.06	9.04	0.0076	0.5816						
	8	9	0.0018	-0.0001	0.0002	0.9791	0.7165	0.1006	0.0001	0.0117	-0.0073	0.5764						
			0.0014	0.0002	0.0002		0.6875	0.0984	-0.0001	0.0113								
339	8	7	0.899	0.03	-0.004	-0.0188	-0.03	9.11	0.06	9.03	0.1281	0.6548						
	8	9	0.0027	-0.0002	0.0000	1.2620	0.7472	0.0528	0.0013	0.0069	0.0688	0.6521						
			0.0009	0.0002	0.0000		0.2635	0.0540	0.0007	0.0070								

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																				
340	1	7	0.900	-2.98	-0.003	-0.00	-0.02	15.07	0.05	15.05	0.1039	0.6506	0.0023	-0.0001	0.0003	-0.2521	-0.6838	0.0582	0.0012	0.0075	0.0563	0.6431
340	2	7	0.898	0.000	0.0004	-0.0004	0.8621	15.09	0.06	15.06	0.0747	0.5998	0.0025	0.0002	0.0000	0.0105	0.790	0.0611	0.0094	0.0094	0.0474	0.5946
340	3	7	0.899	1.10	-0.0004	-0.0001	-0.02	15.09	0.06	15.07	0.0779	0.5916	0.0025	0.0002	0.0004	-0.0063	0.8343	0.0612	0.0095	0.0096	0.0514	0.5821
340	4	7	0.900	3.16	-0.0002	-0.0003	-0.02	15.09	0.06	15.06	0.0452	0.5917	0.0025	0.0002	0.0003	-0.0214	0.6311	0.0607	0.0100	0.0101	0.0227	0.5742
340	5	7	0.898	6.24	-0.0002	-0.0002	-0.02	15.09	0.06	15.05	0.0194	0.5826	0.0018	0.0002	0.0002	-0.1713	0.5498	0.0603	0.0110	0.0109	0.0194	0.5747
340	6	7	0.898	9.33	-0.0002	-0.0003	-0.02	15.09	0.06	15.06	0.0070	0.5847	0.0019	0.0001	0.0002	-0.3147	0.8422	0.0601	0.0120	0.0115	0.0099	0.5716
340	7	7	0.898	12.49	-0.0002	0.0002	-0.02	15.09	0.06	15.06	0.0031	0.5805	0.0017	0.0001	0.0002	-0.3678	0.7249	0.0600	0.0120	0.0119	0.0031	0.5687
340	8	7	0.897	0.02	-0.0003	-0.0003	-0.02	15.09	0.06	15.07	0.0734	0.5963	0.0028	0.0002	0.0000	0.0065	0.7036	0.0611	0.0094	0.0094	0.0734	0.5923
342	3	7	1.251	-3.15	-0.0002	-0.0001	-0.02	-2.98	0.06	-2.98	-0.0488	0.6767	0.0009	0.0002	0.0001	-0.7922	1.1050	0.0004	-0.0057	-0.0058	-0.0168	0.6934
342	4	7	1.250	-0.06	-0.0003	-0.0002	-0.02	-2.95	0.06	-2.97	-0.0535	0.7210	0.0019	0.0001	0.0001	-0.4280	0.7221	0.0601	-0.0020	-0.0021	-0.0528	0.7151
342	5	7	1.250	0.99	-0.0003	-0.0002	-0.02	-2.94	0.06	-2.97	-0.1868	0.7384	0.0020	0.0001	0.0001	-0.3852	0.6780	0.0602	-0.0008	-0.0009	-0.1499	0.7590

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key															
342	6	7	1.248	5.12	-0.001	-0.0001	-0.4872	0.5011	-0.02	-2.72	0.0108	0.0002	0.06	-2.94	0.0014	0.1229	0.6580	
		8	0.0015	-0.0001	0.0002	0.0002	1.3862	0.9239	0.0090	0.0001	0.0001	0.0001	0.0001	0.0001	0.0011	-0.0800	0.6179	
		9	0.0012	0.0003	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	
342	7	7	1.250	6.24	-0.000	-0.0001	-1.0971	0.5820	-0.02	-2.89	0.0368	0.0000	0.06	-2.93	0.0049	0.0022	0.6655	
		8	0.0008	-0.0001	0.0002	0.0002	1.4738	1.0874	0.0368	0.0003	0.0003	0.0003	0.0003	0.0049	0.0049	-0.0443	0.6649	
		9	0.0012	0.0003	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	
342	8	7	1.249	9.37	-0.000	-0.0001	-1.6508	0.6109	-0.02	-2.87	0.0628	0.0003	0.06	-2.91	0.0083	-0.0240	0.6615	
		8	0.0006	-0.0002	0.0002	0.0002	1.3604	1.0634	0.0634	0.0634	0.0634	0.0634	0.0634	0.0634	0.0634	-0.0512	0.6533	
		9	0.0013	0.0003	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	
342	9	7	1.249	12.58	-0.001	-0.0001	-1.5972	1.0963	-0.02	-2.85	0.0863	0.0006	0.06	-2.91	0.0111	-0.0356	0.6446	
		8	0.0006	-0.0001	0.0001	0.0001	3.6383	1.6066	0.0866	0.0866	0.0866	0.0866	0.0866	0.0866	0.0866	-0.0601	0.6407	
		9	0.0005	0.0003	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	
342	10	7	1.250	-0.05	-0.002	-0.0001	-0.4053	0.5960	-0.02	-2.95	0.0145	0.0001	0.06	-2.97	0.0021	-0.0503	0.7308	
		8	0.0021	-0.0001	0.0002	0.0002	1.6359	0.8770	0.0154	0.0154	0.0154	0.0154	0.0154	0.0154	0.0154	0.0545	0.7313	
		9	0.0011	0.0003	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	
343	1	7	1.250	-3.12	-0.001	-0.0001	-0.6070	0.6389	-0.02	-0.09	0.0263	0.0001	0.06	-0.06	0.0036	-0.0316	0.6892	
		8	0.0012	-0.0001	0.0001	0.0001	2.4536	1.1265	0.0261	0.0261	0.0261	0.0261	0.0261	0.0261	0.0261	0.0215	0.7001	
		9	0.0006	0.0003	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	
343	2	7	1.249	-0.02	-0.002	-0.0001	-0.4091	0.5815	-0.02	-0.06	0.0006	0.0002	0.06	-0.04	0.0001	-1.3596	0.7927	
		8	0.0021	-0.0001	0.0001	0.0001	1.6052	0.7298	0.0004	0.0004	0.0004	0.0004	0.0004	0.0004	0.0004	2.6033	1.6443	
		9	0.0011	0.0003	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	
343	3	7	1.251	1.02	-0.002	-0.0001	-0.3741	0.6621	-0.02	-0.05	0.0088	0.0002	0.05	-0.03	0.0011	0.1162	0.6541	
		8	0.0020	-0.0001	0.0001	0.0001	1.7844	0.9251	0.0073	0.0073	0.0073	0.0073	0.0073	0.0073	0.0073	-0.1086	0.6236	
		9	0.0010	0.0003	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	
343	4	7	1.251	3.14	-0.001	-0.0001	-0.4958	0.5227	-0.02	-0.03	0.0268	0.0001	0.06	-0.035	0.0032	-0.0205	0.6629	
		8	0.0014	-0.0001	0.0001	0.0001	1.3766	0.7321	0.0247	0.0247	0.0247	0.0247	0.0247	0.0247	0.0247	0.0545	0.6584	
		9	0.0012	0.0003	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	
343	5	7	1.249	6.25	-0.000	-0.0001	-1.7117	0.7931	-0.02	-0.00	0.0528	0.0002	0.06	-0.00	0.0070	-0.0250	0.6639	
		8	0.0006	-0.0002	0.0002	0.0002	1.3661	0.8021	0.0528	0.0528	0.0528	0.0528	0.0528	0.0528	0.0528	-0.0530	0.6607	
		9	0.0015	0.0004	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	
343	6	7	1.249	9.40	-0.000	-0.0001	-1.3429	0.3646	-0.02	0.01	0.0777	0.0005	0.06	0.0101	-0.0368	0.6510		
		8	0.0007	-0.0002	0.0002	0.0002	1.2777	0.8609	0.0775	0.0775	0.0775	0.0775	0.0775	0.0775	0.0775	-0.0543	0.6490	
		9	0.0015	0.0004	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																					
343	7	1.250	12.64	-0.0001	-0.0001	-1.0221	-0.02	0.986	0.05	0.06	0.0124	-0.0371	0.6314	0.0008	-0.0004	0.0001	0.0001	0.8426	0.0990	-0.0007	0.0124	-0.0616	0.6283
343	8	1.250	-0.0001	0.0004	0.0001	2.2333	0.9571	0.0990	0.05	0.06	0.0124	-0.0371	0.6314	0.0008	0.0004	0.0001	0.0001	0.8426	0.0990	-0.0007	0.0124	-0.0616	0.6283
344	7	1.250	-0.0001	0.0004	0.0001	1.7991	-0.02	-0.0005	-0.07	0.06	-0.0001	-1.3621	1.7215	0.0021	0.0004	0.0001	0.0001	0.5230	-0.0005	0.0001	-0.0001	-2.1103	1.7583
344	8	1.249	-3.11	-0.0001	-0.0002	-0.8280	-0.02	0.98	0.212	0.06	0.94	-0.0453	0.6727	0.0009	-0.0001	0.0001	0.0001	1.142	-0.0212	0.0001	-0.0028	-0.0355	0.6792
344	9	0.0011	0.0002	0.0001	0.0001	0.8793	0.5780	-0.0207	-0.98	0.06	0.0028	-0.0453	0.6792	0.0011	0.0002	0.0001	0.0001	0.5780	-0.0207	0.0001	-0.0028	-0.0355	0.6792
344	7	1.248	-0.0001	0.0002	-0.0003	-0.3338	-0.02	1.01	1.01	0.06	0.96	-0.2444	0.7415	0.0020	0.0001	0.0001	0.0001	0.7612	0.0045	0.0002	0.0006	0.2444	0.7415
344	8	0.0020	-0.0001	0.0002	0.0001	1.0020	0.7056	0.0045	0.0045	0.06	0.0006	-0.2444	0.7415	0.0013	0.0002	0.0001	0.0001	0.7056	0.0045	0.0002	0.0006	0.2444	0.7415
344	9	0.0015	0.0002	0.0001	0.0001	0.8770	0.5953	0.0129	0.0129	0.06	0.0017	-0.0737	0.6876	0.0015	0.0002	0.0001	0.0001	0.5953	0.0129	0.0002	0.0017	-0.0737	0.6876
344	7	1.247	3.14	-0.0001	-0.0002	-0.4198	-0.02	1.04	1.04	0.06	0.97	-0.0316	0.6934	0.0017	0.0001	0.0001	0.0001	0.5184	0.0314	0.0001	0.0043	0.0316	0.6934
344	8	0.0017	-0.0001	0.0002	0.0001	0.8824	0.6990	0.0299	0.0299	0.06	0.0041	-0.0523	0.6851	0.0015	0.0002	0.0001	0.0001	0.6990	0.0299	0.0003	0.0041	-0.0523	0.6851
344	9	0.0015	0.0003	0.0001	0.0001	0.9906	0.9051	0.0580	0.0580	0.06	0.0077	-0.0225	0.6689	0.0015	0.0003	0.0001	0.0001	0.9051	0.0580	0.0005	0.0077	-0.0225	0.6689
344	7	1.248	9.41	-0.0002	-0.0003	-2.4711	-0.02	1.09	1.09	0.06	1.00	-0.0315	0.6528	0.0004	0.0002	0.0001	0.0001	1.6252	0.0820	0.0005	0.0107	-0.0315	0.6528
344	8	0.0004	-0.0002	0.0003	0.0001	1.0092	0.8472	0.0824	0.0824	0.06	0.0107	-0.0554	0.6501	0.0016	0.0003	0.0001	0.0001	0.8472	0.0824	0.0009	0.0107	-0.0554	0.6501
344	9	0.0006	0.0003	0.0001	0.0001	2.8103	1.5218	0.1029	0.1029	0.06	0.0129	-0.0604	0.6265	0.0006	0.0003	0.0001	0.0001	1.5218	0.1029	0.0012	0.0129	-0.0604	0.6265
344	7	1.250	-0.0001	0.0003	-0.0001	-0.3911	-0.02	1.01	1.01	0.06	0.96	-0.1939	0.6373	0.0018	-0.0001	0.0001	0.0001	0.7795	0.0045	0.0001	0.0005	0.1939	0.6373
344	8	0.0018	-0.0001	0.0003	0.0001	1.4332	0.7995	0.0044	0.0044	0.06	0.0005	-0.2173	0.5700	0.0012	0.0003	0.0001	0.0001	0.7995	0.0044	0.0001	0.0005	-0.2173	0.5700
344	9	0.0012	0.0003	0.0001	0.0001	2.1422	0.9392	-0.0096	-0.0096	0.06	0.0013	-0.0351	0.7169	0.0012	0.0003	0.0001	0.0001	0.9392	-0.0096	0.0002	0.0013	-0.0351	0.7169
345	7	1.250	-3.11	-0.0001	-0.0003	-0.4731	-0.02	2.95	2.95	0.06	3.03	-0.0351	0.7169	0.0017	-0.0001	0.0001	0.0001	0.4252	-0.0102	0.0000	-0.0014	-0.0351	0.7169
345	8	0.0017	-0.0001	0.0003	0.0001	2.1422	0.9392	-0.0096	-0.0096	0.06	0.0013	-0.0351	0.7169	0.0007	-0.0003	0.0001	0.0001	0.4252	-0.0102	0.0000	-0.0014	-0.0351	0.7169
345	9	0.0007	0.0003	0.0001	0.0001	2.1422	0.9392	-0.0096	-0.0096	0.06	0.0013	-0.0351	0.7169	0.0007	0.0003	0.0001	0.0001	0.4252	-0.0102	0.0000	-0.0014	-0.0351	0.7169

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																						
345	2	1.248	-0.0019	-0.0010	-0.0003	-0.0001	-0.0002	-0.0001	-0.3887	1.5619	0.6646	0.8925	2.98	0.0147	0.0150	0.0001	0.0002	3.06	0.0020	0.0020	0.0611	0.0746	0.6990	0.6748
345	3	1.248	-0.0017	-0.0012	0.0003	0.0001	0.0002	0.0002	-0.3732	1.4671	0.7953	0.8260	3.00	0.0252	0.0243	0.0001	0.0002	3.06	0.0034	0.0032	0.0210	0.0574	0.6753	0.6757
345	4	1.247	-0.0011	-0.0013	0.0003	0.0001	0.0001	0.0001	-0.5856	1.1377	0.8048	0.7234	3.01	0.0431	0.0416	0.0000	0.0004	3.08	0.0058	0.0056	0.0064	0.0551	0.6766	0.6767
345	5	1.247	-0.0004	-0.0016	0.0003	0.0001	0.0002	0.0001	-2.0576	1.0972	1.7110	0.7333	3.03	0.0683	0.0688	0.0004	0.0007	3.10	0.0090	0.0090	0.0323	0.0558	0.6630	0.6567
345	6	1.248	-0.0003	-0.0014	0.0003	0.0001	0.0002	0.0001	-3.2788	1.1263	1.5720	0.8279	3.06	0.0911	0.0920	0.0006	0.0010	3.11	0.0117	0.0117	0.0353	0.0591	0.6440	0.6407
345	7	1.249	-0.0004	-0.0005	0.0003	0.0001	0.0001	0.0001	-1.8739	3.1957	1.7908	1.5032	3.07	0.1097	0.1108	0.0008	0.0013	3.12	0.0137	0.0137	0.0385	0.0620	0.6254	0.6219
345	8	1.250	-0.0020	-0.0012	0.0003	0.0001	0.0002	0.0001	-0.4240	1.5794	0.6481	0.8283	2.98	0.0149	0.0148	0.0001	0.0002	3.07	0.0020	0.0019	0.0488	0.0756	0.6762	0.6694
346	1	1.249	-0.0011	-0.0008	0.0003	0.0001	0.0001	0.0001	-0.6488	1.8092	0.7509	0.6748	6.01	0.0043	0.0048	0.0001	0.0002	6.00	0.0006	0.0006	0.1261	0.2434	0.7367	0.7150
346	2	1.249	-0.0020	-0.0011	0.0003	0.0001	0.0002	0.0001	-0.4677	1.6821	0.6214	0.8046	6.03	0.0317	0.0315	0.0000	0.0003	6.01	0.0043	0.0043	0.0003	0.0619	0.6871	0.6871
346	3	1.250	-0.0018	-0.0012	0.0003	0.0001	0.0002	0.0001	-0.4045	1.3501	0.6859	0.6306	6.04	0.0412	0.0402	0.0000	0.0004	6.02	0.0056	0.0054	0.0070	0.0598	0.6884	0.6819
346	4	1.248	-0.0010	-0.0012	0.0003	0.0001	0.0001	0.0001	-0.6866	1.3711	0.8236	0.7392	6.06	0.0599	0.0578	0.0003	0.0006	6.03	0.0079	0.0077	0.0274	0.0594	0.6666	0.6709

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key														
346	5	7	1.249	0.0004	0.0017	6.31	-0.0001	-2.0982	-0.02	0.0829	6.08	0.0006	0.0108	-0.0369	0.6528		
		8	0.0004	0.0003	0.0003	0.0001	0.0002	1.1311	1.5785	0.0832	0.0832	-0.0009	0.0108	-0.0575	0.6494		
		9	0.0017	0.0003	0.0003	0.0001	0.0002	1.1311	0.7539	0.0832	0.0832	-0.0009	0.0108	-0.0575	0.6494		
346	6	7	1.249	0.0000	0.0015	9.44	-0.0001	-12.1431	-0.02	6.10	6.10	0.0007	6.06	-0.0343	0.6309		
		8	0.0000	0.0003	0.0003	0.0001	0.0002	1.1611	7.5348	0.1028	0.1028	-0.0012	0.0129	-0.0585	0.6256		
		9	0.0015	0.0003	0.0003	0.0001	0.0002	1.1611	0.9018	0.1038	0.1038	-0.0012	0.0129	-0.0585	0.6256		
346	7	7	1.248	0.0008	0.0008	12.67	-0.0001	-1.1136	-0.02	6.11	6.11	0.0007	6.06	-0.0427	0.6193		
		8	0.0008	0.0003	0.0003	0.0001	0.0001	2.3458	0.7179	0.1202	0.1202	-0.0010	0.0148	-0.0652	0.6135		
		9	0.0008	0.0003	0.0003	0.0001	0.0001	2.3458	0.8191	0.1209	0.1209	-0.0015	0.0148	-0.0652	0.6135		
346	8	7	1.249	0.0019	0.0010	0.0003	-0.0002	-0.4162	-0.02	6.03	6.03	0.0000	6.02	-0.0054	0.6772		
		8	0.0019	0.0003	0.0003	0.0001	0.0001	1.9098	0.6877	0.0319	0.0319	-0.0004	0.0042	-0.0635	0.6806		
		9	0.0010	0.0003	0.0003	0.0001	0.0001	1.9098	0.8934	0.0315	0.0315	-0.0004	0.0042	-0.0635	0.6806		
347	1	7	1.249	0.0011	0.0008	5.04	-0.0002	-0.7039	-0.02	9.05	9.05	0.0000	8.98	0.0116	0.7172		
		8	0.0011	0.0003	0.0003	0.0001	0.0001	1.8605	0.9384	0.0199	0.0199	0.0000	0.0028	-0.0808	0.6942		
		9	0.0008	0.0003	0.0003	0.0001	0.0001	1.8605	0.7998	0.0202	0.0202	-0.0003	0.0028	-0.0808	0.6942		
347	2	7	1.250	0.0019	0.0008	0.04	-0.0002	-0.4410	-0.02	9.08	9.08	0.0002	9.00	-0.0272	0.6792		
		8	0.0019	0.0003	0.0003	0.0001	0.0001	2.1547	0.6323	0.0486	0.0486	-0.0006	0.0065	-0.0643	0.6828		
		9	0.0008	0.0003	0.0003	0.0001	0.0001	2.1547	1.0628	0.0477	0.0477	-0.0006	0.0065	-0.0643	0.6828		
347	3	7	1.249	0.0020	0.0011	1.10	-0.0002	-0.4170	-0.02	9.09	9.09	0.0003	9.00	-0.0286	0.6752		
		8	0.0020	0.0003	0.0003	0.0001	0.0001	1.5587	0.6012	0.0579	0.0579	-0.0003	0.0075	-0.0617	0.6750		
		9	0.0011	0.0003	0.0003	0.0001	0.0001	1.5587	0.7929	0.0560	0.0560	-0.0006	0.0075	-0.0617	0.6750		
347	4	7	1.250	0.0010	0.0011	3.19	-0.0001	-0.6130	-0.02	9.11	9.11	0.0005	9.01	-0.0370	0.6567		
		8	0.0010	0.0003	0.0003	0.0001	0.0001	1.6365	0.7133	0.0755	0.0755	-0.0008	0.0096	-0.0605	0.6609		
		9	0.0011	0.0003	0.0003	0.0001	0.0001	1.6365	0.9247	0.0727	0.0727	-0.0008	0.0096	-0.0605	0.6609		
347	5	7	1.248	0.0004	0.0015	6.32	-0.0001	-2.1621	-0.02	9.13	9.13	0.0006	9.02	-0.0381	0.6396		
		8	0.0004	0.0003	0.0003	0.0001	0.0001	1.2351	1.3349	0.0964	0.0964	-0.0007	0.0123	-0.0592	0.6384		
		9	0.0015	0.0003	0.0003	0.0001	0.0001	1.2351	0.8067	0.0963	0.0963	-0.0011	0.0123	-0.0592	0.6384		
347	6	7	1.248	0.0005	0.0016	9.46	-0.0002	-1.6666	-0.02	9.15	9.15	0.0007	9.05	-0.0385	0.6255		
		8	0.0005	0.0003	0.0003	0.0001	0.0001	1.1735	0.2341	0.1141	0.1141	-0.0008	0.0142	-0.0585	0.6211		
		9	0.0016	0.0003	0.0003	0.0001	0.0001	1.1735	0.7336	0.1145	0.1145	-0.0013	0.0142	-0.0585	0.6211		
347	7	7	1.248	0.0004	0.0005	12.69	-0.0001	-1.9080	-0.02	9.16	9.16	0.0006	9.05	-0.0448	0.6137		
		8	0.0004	0.0001	0.0004	0.0001	0.0001	3.4642	1.3909	0.1290	0.1290	-0.0011	0.0158	-0.0637	0.6106		
		9	0.0005	0.0004	0.0004	0.0001	0.0001	3.4642	1.4512	0.1288	0.1288	-0.0016	0.0157	-0.0637	0.6106		

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

← See Key →

Run	Pt.	Cd																	
347	0	7	1.248	0.0015	0.0007	0.04	-0.0001	-0.0002	-0.5505	1.2378	0.9145	9.08	0.0484	-0.0002	0.05	9.00	0.0065	-0.0270	0.6791
		6	0.0007	-0.0003			0.0001	0.0001	2.4139	1.2378	0.9145	9.08	0.0484	-0.0006	0.0006	0.0064	0.0064	-0.0639	0.6832
		9	0.0007	0.0003			0.0001	0.0001	2.4139	1.2378	0.9145	9.08	0.0484	-0.0006	0.0006	0.0064	0.0064	-0.0639	0.6832
348	1	7	1.246	0.0011	0.0006	-2.99	-0.0001	-0.0001	-0.7641	1.1786	0.8640	15.04	0.0519	-0.0003	0.05	15.01	0.0072	-0.0375	0.6942
		8	0.0011	-0.0001			0.0001	0.0001	2.7582	1.1786	0.8640	15.04	0.0519	-0.0007	0.0007	0.0072	0.0072	-0.0739	0.6826
		9	0.0006	0.0003			0.0001	0.0001	2.7582	1.1786	0.8640	15.04	0.0519	-0.0007	0.0007	0.0072	0.0072	-0.0739	0.6826
349	2	7	1.251	0.0018	0.0011	0.07	-0.0002	-0.0001	-0.4459	0.7205	0.6904	15.07	0.0786	-0.0006	0.05	15.03	0.0104	-0.0392	0.6627
		6	0.0018	-0.0001			0.0001	0.0001	1.5544	0.7205	0.6904	15.07	0.0786	-0.0010	0.0010	0.0104	0.0104	-0.0639	0.6592
		9	0.0011	0.0003			0.0001	0.0001	1.5544	0.7205	0.6904	15.07	0.0786	-0.0010	0.0010	0.0103	0.0103	-0.0639	0.6592
348	3	7	1.251	0.0017	0.0011	1.13	-0.0002	-0.0001	-0.4795	0.6738	0.6738	15.08	0.0864	-0.0006	0.05	15.03	0.0113	-0.0360	0.6568
		6	0.0017	-0.0001			0.0001	0.0001	1.6767	0.6738	0.6738	15.08	0.0864	-0.0010	0.0010	0.0111	0.0111	-0.0624	0.6497
		9	0.0011	0.0003			0.0001	0.0001	1.6767	0.6738	0.6738	15.08	0.0864	-0.0010	0.0010	0.0111	0.0111	-0.0624	0.6497
348	4	7	1.250	0.0009	0.0010	3.23	-0.0001	-0.0002	-0.7055	0.9227	0.7089	15.09	0.1014	-0.0007	0.05	15.05	0.0129	-0.0376	0.6364
		6	0.0009	-0.0001			0.0001	0.0002	1.8344	0.9227	0.7089	15.09	0.1014	-0.0012	0.0012	0.0129	0.0129	-0.0618	0.6332
		9	0.0010	0.0004			0.0002	0.0002	1.8344	0.9227	0.7089	15.09	0.1014	-0.0012	0.0012	0.0126	0.0126	-0.0618	0.6332
348	5	7	1.249	0.0006	0.0009	6.37	-0.0003	-0.0003	-1.2582	1.5716	0.6952	15.11	0.1179	-0.0013	0.05	15.06	0.0145	-0.0363	0.6283
		8	0.0006	-0.0001			0.0003	0.0003	2.0032	1.5716	0.6952	15.11	0.1179	-0.0013	0.0013	0.0145	0.0145	-0.0561	0.6146
		9	0.0009	0.0003			0.0003	0.0003	2.0032	1.5716	0.6952	15.11	0.1179	-0.0013	0.0013	0.0145	0.0145	-0.0561	0.6146
348	6	7	1.248	0.0002	0.0015	9.47	-0.0003	-0.0002	-3.5101	1.3252	0.8540	15.12	0.1320	-0.0011	0.05	15.06	0.0162	-0.0453	0.6140
		6	0.0002	-0.0001			0.0002	0.0002	1.1983	1.3252	0.8540	15.12	0.1320	-0.0016	0.0016	0.0162	0.0162	-0.0617	0.6051
		9	0.0015	0.0003			0.0002	0.0002	1.1983	1.3252	0.8540	15.12	0.1320	-0.0016	0.0016	0.0160	0.0160	-0.0617	0.6051
348	7	7	1.248	0.0006	0.0006	12.68	-0.0001	-0.0001	-1.3553	1.1812	0.812	15.12	0.1343	-0.0011	0.05	15.06	0.0159	-0.0418	0.5978
		8	0.0006	-0.0001			0.0001	0.0001	3.1725	1.1812	0.812	15.12	0.1343	-0.0016	0.0016	0.0159	0.0159	-0.0617	0.5951
		9	0.0006	0.0003			0.0001	0.0001	3.1725	1.1812	0.812	15.12	0.1343	-0.0016	0.0016	0.0159	0.0159	-0.0617	0.5951
348	8	7	1.248	0.0018	0.0008	0.07	-0.0002	-0.0001	-0.4798	0.9351	0.6823	15.07	0.0786	-0.0006	0.05	15.03	0.0104	-0.0399	0.6602
		6	0.0018	-0.0001			0.0001	0.0001	2.2590	0.9351	0.6823	15.07	0.0786	-0.0010	0.0010	0.0104	0.0104	-0.0640	0.6570
		9	0.0008	0.0004			0.0001	0.0001	2.2590	0.9351	0.6823	15.07	0.0786	-0.0010	0.0010	0.0103	0.0103	-0.0640	0.6570
349	3	7	0.897	-0.0025	-0.0000	-3.09	-0.0003	-0.0003	-0.2084	0.7452	-0.0487	-2.98	-0.0487	-0.0006	0.06	-3.00	-0.0064	0.0717	0.6594
		6	-0.0025	0.0001			0.0003	0.0003	4.2716	10.2369	-0.0487	-2.98	-0.0487	-0.0011	0.0011	-0.0064	-0.0064	0.1276	0.6594
		9	-0.0000	0.0002			-0.0000	-0.0000	4.2716	10.2369	-0.0487	-2.98	-0.0487	-0.0011	0.0011	-0.0059	-0.0059	0.1276	0.6594
349	4	7	0.898	-0.0026	-0.0007	-0.03	-0.0004	-0.0004	-0.0921	0.9140	-0.0140	-2.95	-0.0163	-0.0000	0.07	-2.97	-0.0021	0.0282	0.6516
		6	-0.0026	0.0000			0.0004	0.0004	1.9141	-0.1307	-0.0140	-2.95	-0.0163	-0.0000	0.0000	-0.0021	-0.0021	0.1887	0.6516
		9	-0.0007	0.0002			-0.0000	-0.0000	1.9141	-0.1307	-0.0140	-2.95	-0.0163	-0.0000	0.0000	-0.0021	-0.0021	0.1887	0.6516

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																			
349	5	7 6 9	0.898 0.0026 0.0005	0.0000 0.0000 0.0000	-0.004 0.0000 0.0000	-0.0334 -0.2622 0.0000	-0.02 0.3756 0.0000	-2.93 -0.0061 -0.0062	0.07 0.0000 -0.0003	-2.96 -0.0008 -0.0008	-0.0564 0.2913 0.0000	0.6857 0.7026 0.0000									
349	6	7 6 9	0.899 0.0024 0.0006	0.0000 0.0000 0.0000	-0.003 0.0000 0.0000	-0.0297 1.8617 0.0000	-0.02 0.7985 0.4465	-2.91 0.0113 0.0104	0.06 0.0006 0.0001	-2.95 0.0015 0.0014	0.3039 0.0747 0.0000	0.6993 0.6882 0.0000									
349	7	7 6 9	0.896 0.0016 0.0007	0.0000 0.0000 0.0000	-0.003 0.0002 0.0000	-0.2530 1.7173 0.0000	-0.02 0.9708 1.3400	-2.87 0.0431 0.0463	0.06 0.0012 0.0005	-2.91 0.0057 0.0062	0.1478 0.0612 0.0000	0.6681 0.6699 0.0000									
349	8	7 6 9	0.897 0.0018 0.0014	0.0001 0.0003 0.0000	-0.002 0.0002 0.0000	-0.3335 1.1299 0.0000	-0.02 0.7787 0.8894	-2.85 0.0681 0.0694	0.07 0.0013 0.0009	-2.89 0.0081 0.0083	0.1020 0.0697 0.0000	0.6002 0.6045 0.0000									
349	9	7 6 9	0.899 0.0017 0.0003	0.0001 0.0002 0.0000	-0.003 0.0002 0.0000	-0.4456 3.4950 0.0000	-0.02 0.8953 3.0186	-2.83 0.0850 0.0868	0.06 0.0012 0.0008	-2.89 0.0098 0.0100	0.0722 0.0464 0.0000	0.5810 0.5778 0.0000									
349	10	7 6 9	0.894 0.0032 0.0005	0.0000 0.0003 0.0000	-0.004 0.0000 0.0000	-0.0607 3.3440 0.0000	-0.02 0.6602 0.2854	-2.95 -0.0165 -0.0165	0.06 -0.0001 -0.0006	-2.99 -0.0023 -0.0023	0.0478 0.1955 0.0000	0.6843 0.7001 0.0000									
350	1	7 6 9	0.899 0.0023 -0.0001	0.0001 0.0002 0.0000	-0.003 0.0000 -0.0000	-0.2380 -10.4447 0.0000	-0.02 0.8240 1.1589	-0.11 -0.0316 -0.0317	0.06 -0.0003 -0.0007	-0.06 -0.0043 -0.0043	0.0531 0.1108 0.0000	0.6829 0.6794 0.0000									
350	2	7 6 9	0.900 0.0026 0.0008	0.0000 0.0000 0.0000	-0.004 0.0000 0.0000	-0.0566 1.7066 0.0000	-0.02 0.8640 0.0370	-0.06 -0.0009 0.0018	0.06 0.0003 -0.0002	-0.02 0.0000 0.0000	-1.8341 -0.6667 0.0000	0.4406 0.7375 0.0000									
350	3	7 6 9	0.899 0.0024 0.0006	0.0000 0.0000 0.0000	-0.004 0.0000 0.0000	-0.1011 2.0045 0.0000	-0.02 0.9901 0.2588	-0.05 0.0076 0.0093	0.06 0.0006 0.0001	-0.01 0.0011 0.0013	0.4433 0.0608 0.0000	0.7734 0.6987 0.0000									
350	4	7 6 9	0.899 0.0021 0.0006	0.0001 0.0000 0.0000	-0.004 0.0000 0.0000	-0.0212 2.0598 0.0000	-0.02 0.9902 0.7455	-0.03 0.0328 0.0321	0.06 0.0008 0.0002	-0.00 0.0045 0.0043	0.1343 0.0404 0.0000	0.6862 0.6845 0.0000									
350	5	7 6 9	0.900 0.0015 0.0012	0.0000 0.0000 0.0000	-0.003 0.0003 0.0001	-0.2343 1.2400 0.0000	-0.02 0.0192 0.7219	0.00 0.0586 0.0600	0.07 0.0013 0.0009	0.02 0.0071 0.0075	0.1168 0.0768 0.0000	0.6139 0.6265 0.0000									

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd																			
350	6	7	0.899	7.28	-0.003	-0.003	-0.02	0.01	0.06	0.04	0.0790	0.5880	0.0015	-0.001	0.0003	0.9608	0.0795	0.0012	0.0093	0.0542	0.5865
			0.0013	0.0003	0.0002	1.1535	1.0229	0.0811	0.0008	0.0095											
350	7	7	0.900	12.46	-0.002	-0.002	-0.02	0.01	0.06	0.03	0.0389	0.5834	0.0016	-0.001	0.0002	0.8197	0.0898	0.0007	0.0104	0.0167	0.5754
			0.0009	0.0003	0.0002	1.5749	1.0841	0.0915	0.0003	0.0105											
350	8	7	0.899	-0.003	-0.005	-0.0680	-0.02	-0.07	0.06	-0.003	-2.2128	0.7287	0.0025	-0.000	0.0000	0.9780	-0.0018	0.0002	0.0002	-0.6717	0.6745
			0.0002	0.0003	0.0000	7.3149	1.9580	0.0018	-0.0002	-0.0002											
351	1	7	0.898	-3.07	-0.003	-0.2510	-0.02	0.98	0.06	0.92	0.0657	0.6968	0.0024	-0.000	0.0000	0.7759	-0.0003	-0.0032	0.1409	0.6677	
			-0.0003	0.0002	-0.0000	-4.0500	0.1546	-0.0243	-0.0006	-0.0006											
351	2	7	0.898	-0.00	-0.004	-0.0422	-0.02	1.03	0.06	0.96	0.9141	1.0046	0.0027	-0.000	0.0000	0.8615	0.0006	0.0007	0.0124	0.6668	
			0.0005	0.0002	0.0000	2.5303	0.5293	0.0064	0.0000	0.0008											
351	3	7	0.898	1.03	-0.004	-0.0395	-0.02	1.04	0.06	0.97	0.2495	0.7368	0.0030	-0.000	0.0000	0.7458	0.0168	0.0000	0.0222	0.0215	0.6689
			0.0005	0.0002	0.0000	2.7037	0.8261	0.0168	0.0000	0.0022											
351	4	7	0.898	3.13	-0.004	0.0222	-0.02	1.06	0.06	0.98	0.1333	0.6758	0.0021	-0.000	0.0000	0.9707	0.0389	0.0010	0.0052	0.0462	0.6814
			0.0006	0.0002	0.0001	2.1122	0.7907	0.0389	0.0003	0.0053											
351	5	7	0.897	6.22	-0.003	-0.1688	-0.02	1.09	0.06	1.01	0.1158	0.6169	0.0018	-0.000	0.0000	0.8701	0.0644	0.0009	0.0079	0.0756	0.6146
			0.0009	0.0002	0.0002	1.5828	1.2156	0.0644	0.0004	0.0079											
351	6	7	0.899	9.30	-0.003	-0.3164	-0.02	1.10	0.06	1.02	0.0833	0.5889	0.0018	-0.000	0.0000	0.8559	0.013	0.0096	0.0096	0.0562	0.5794
			0.0014	0.0003	0.0002	1.0996	1.0108	0.0832	0.0009	0.0096											
351	7	7	0.897	12.45	-0.003	-0.4216	-0.02	1.10	0.07	1.02	0.0290	0.5771	0.0016	-0.000	0.0000	0.9558	0.0005	0.0104	0.0123	0.5771	
			0.0011	0.0003	0.0002	1.3887	0.9180	0.0912	0.0002	0.0104											
351	8	7	0.899	-0.00	-0.004	-0.0538	-0.02	1.02	0.06	0.96	0.4789	0.6259	0.0025	-0.000	0.0000	0.9704	0.0050	0.0006	0.0257	0.6259	
			0.0000	0.0002	0.0001	55.6982	18.9771	0.0065	0.0000	0.0009											

See Key

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key												
352	1	7	0.898	-3.05	-0.0001	-0.00	-0.00	-0.02	0.7196	2.96	0.06	3.03	0.1044	0.6943
	8	8	0.0025	-0.0001	0.0000	0.0003	0.0000	0.7196	-0.0090	-0.0090	-0.0001	-0.0012	0.2954	0.6779
	9	9	-0.0002	0.0002	-0.0000	-0.0000	0.0250	-0.0084	-0.0004	-0.0004	-0.0011	0.0011		
352	2	7	0.897	-0.00	-0.0000	-0.00	-0.02	0.9242	0.185	3.00	0.06	3.06	0.1945	0.7206
	8	8	0.0025	-0.0000	0.0000	0.0004	0.9242	0.0215	0.0007	0.0007	0.0026	0.0026	0.0244	0.6907
	9	9	0.0004	0.0002	0.0000	0.0000	0.7677	0.0215	0.0001	0.0001	0.0029	0.0029		
352	3	7	0.896	1.02	-0.0000	-0.00	-0.02	0.6825	3.01	0.06	3.07	0.1298	0.6956	
	8	8	0.0031	0.0000	0.0000	0.0004	0.6825	0.308	0.0008	0.0008	0.0042	0.0042	0.0297	0.6868
	9	9	0.0008	0.0002	0.0000	0.0000	0.3817	0.0322	0.0001	0.0001	0.0044	0.0044		
352	4	7	0.898	3.11	-0.0000	-0.00	-0.02	0.9472	3.04	0.07	3.09	0.1253	0.6313	
	8	8	0.0020	-0.0000	0.0000	0.0003	0.9472	0.0492	0.0012	0.0012	0.0062	0.0062	0.0637	0.6589
	9	9	0.0007	0.0002	0.0000	0.0000	0.6708	0.0506	0.0006	0.0006	0.0066	0.0066		
352	5	7	0.898	6.20	-0.0001	-0.00	-0.02	0.9965	3.06	0.06	3.11	0.0914	0.5978	
	8	8	0.0015	-0.0001	0.0000	0.0003	0.9965	0.0730	0.0013	0.0013	0.0088	0.0088	0.0648	0.6030
	9	9	0.0010	0.0003	0.0000	0.0002	1.0453	0.0730	0.0009	0.0009	0.0088	0.0088		
352	6	7	0.898	9.30	-0.0001	-0.00	-0.02	0.9566	3.06	0.06	3.11	0.0551	0.5791	
	8	8	0.0015	-0.0001	0.0000	0.0003	0.9566	0.0849	0.0009	0.0009	0.0098	0.0098	0.0360	0.5754
	9	9	0.0015	0.0003	0.0000	0.0003	1.0851	0.0876	0.0006	0.0006	0.0100	0.0100	0.0231	0.5744
352	7	7	0.898	12.44	-0.0001	-0.00	-0.02	0.6946	3.06	0.06	3.11	0.0231	0.5799	
	8	8	0.0015	-0.0001	0.0000	0.0002	0.6946	0.0936	0.0004	0.0004	0.0108	0.0108	0.0071	0.5744
	9	9	0.0009	0.0003	0.0000	0.0001	1.0700	0.0940	0.0001	0.0001	0.0108	0.0108	0.0231	0.5744
352	8	7	0.898	-0.02	-0.0000	-0.00	-0.02	0.8804	3.00	0.06	3.05	0.1886	0.7128	
	8	8	0.0027	-0.0000	0.0000	0.0004	0.8804	0.185	0.0006	0.0006	0.0026	0.0026	0.0242	0.7033
	9	9	0.0001	0.0002	0.0000	0.0001	2.9683	0.0210	0.0001	0.0001	0.0029	0.0029	0.0242	0.7033
353	1	7	0.898	-3.05	-0.0000	-0.00	-0.02	0.6869	6.00	0.06	6.00	0.6375	0.8740	
	8	8	0.0027	-0.0000	0.0000	0.0003	0.6869	0.0045	0.0005	0.0005	0.0007	0.0007	0.0367	0.7439
	9	9	-0.0000	0.0002	-0.0000	-0.0000	8.3925	0.0078	0.0000	0.0000	0.0011	0.0011	0.0367	0.7439
353	2	7	0.898	-0.02	-0.0000	-0.00	-0.02	0.6563	6.04	0.06	6.02	0.1065	0.6694	
	8	8	0.0031	-0.0000	0.0000	0.0004	0.6563	0.0393	0.0008	0.0008	0.0052	0.0052	0.0301	0.6874
	9	9	0.0006	0.0002	0.0000	0.0000	0.2685	0.0415	0.0002	0.0002	0.0057	0.0057	0.0301	0.6874
353	3	7	0.897	1.03	-0.0000	-0.00	-0.02	0.7247	6.05	0.06	6.03	0.1256	0.6470	
	8	8	0.0029	-0.0000	0.0000	0.0004	0.7247	0.0470	0.0011	0.0011	0.0065	0.0065	0.0474	0.6611
	9	9	0.0009	0.0002	0.0000	0.0000	0.2643	0.0498	0.0004	0.0004	0.0065	0.0065	0.0474	0.6611

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key									
353	4	0.899	5.13	-0.00	0.0263	-0.02	6.06	0.06	6.05	0.1057	0.6200
	8	0.0019	0.0000	0.0003	0.0153	0.9153	0.0647	0.0013	0.0080	0.0634	0.6240
	9	0.0005	0.0002	0.0000	2.4611	0.8488	0.0650	0.0008	0.0081		
353	5	0.901	6.22	-0.00	-0.2673	-0.02	6.08	0.07	6.06	0.0859	0.5911
	8	0.0014	-0.0000	0.0003	1.3242	1.0931	0.0811	0.0013	0.0095	0.0577	0.5812
	9	0.0009	0.0002	0.0002	-0.9707	1.4017	0.0830	0.0009	0.0096		
353	6	0.899	9.28	-0.00	-0.2180	-0.02	6.07	0.06	6.06	0.0361	0.5809
	8	0.0015	-0.0000	0.0002	0.9707	0.9147	0.0893	0.0006	0.0103	0.0153	0.5743
	9	0.0015	0.0003	0.0003	1.0397	1.0397	0.0905	0.0002	0.0104		
353	7	0.900	12.46	-0.00	-0.4230	-0.02	6.08	0.06	6.06	0.0154	0.5823
	8	0.0016	-0.0001	0.0002	1.9662	0.7799	0.0979	0.0003	0.0114	0.0017	0.5724
	9	0.0006	0.0002	0.0002	1.9662	1.9970	0.0970	0.0000	0.0111		
353	8	0.899	-0.00	-0.00	-0.0579	-0.02	6.04	0.06	6.03	0.1130	0.6780
	8	0.0026	-0.0000	0.0004	4.4974	0.8282	0.0396	0.0008	0.0053	0.0270	0.6854
	9	0.0003	0.0002	0.0000	-0.9707	1.3484	0.0422	0.0002	0.0057		
354	1	0.899	-3.04	-0.00	-0.2519	-0.02	9.06	0.06	9.00	0.1391	0.7353
	8	0.0025	-0.0001	0.0003	13.9371	0.7681	0.0261	0.0007	0.0038	0.0081	0.7115
	9	-0.0000	0.0002	-0.0000	-13.9371	0.2759	0.0295	0.0000	0.0042		
354	2	0.897	0.01	-0.00	-0.0034	-0.02	9.10	0.06	9.02	0.1166	0.6418
	8	0.0025	-0.0000	0.0004	6.3140	0.9091	0.0540	0.0012	0.0069	0.0573	0.6437
	9	0.0001	0.0002	0.0000	-0.9707	2.4655	0.0563	0.0006	0.0072		
354	3	0.900	1.05	-0.00	-0.0049	-0.02	9.11	0.06	9.03	0.1056	0.6278
	8	0.0025	-0.0000	0.0004	1.9345	0.8716	0.0626	0.0013	0.0078	0.0599	0.6318
	9	0.0006	0.0002	0.0001	-0.9707	0.9803	0.0636	0.0007	0.0080		
354	4	0.899	3.13	-0.00	0.0696	-0.02	9.12	0.06	9.04	0.0876	0.6035
	8	0.0021	0.0000	0.0003	1.8119	0.8117	0.0770	0.0013	0.0093	0.0516	0.6022
	9	0.0007	0.0002	0.0001	-0.9707	0.9898	0.0764	0.0007	0.0092		
354	5	0.900	6.22	-0.00	-0.1883	-0.02	9.12	0.06	9.05	0.0499	0.5835
	8	0.0016	-0.0000	0.0002	0.9672	0.8495	0.0861	0.0008	0.0100	0.0333	0.5783
	9	0.0015	0.0003	0.0002	-0.9707	0.7491	0.0883	0.0005	0.0102		
354	6	0.898	9.30	-0.00	-0.3060	-0.02	9.12	0.06	9.04	0.0198	0.5730
	8	0.0015	-0.0000	0.0002	1.1406	0.9220	0.0946	0.0003	0.0108	0.0057	0.5756
	9	0.0012	0.0002	0.0003	-0.9707	1.4953	0.0944	0.0001	0.0108		

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																		
354	7	0.898	12.46	-0.00	-0.3026	-0.02	9.12	0.06	9.05	0.0030	0.5751	0.0021	-0.0001	0.0002	-0.5750	0.1028	0.0000	0.0116	0.0088	0.5731
	8	0.900	0.03	-0.00	-0.0262	-0.02	9.10	0.06	9.03	0.1081	0.6321	0.0030	-0.0000	0.0004	0.7042	0.0544	0.0011	0.0068	0.0554	0.6437
355	1	0.897	-3.00	-0.00	-0.1873	-0.02	15.07	0.06	15.05	0.0943	0.6362	0.0025	-0.0000	0.0003	0.6956	0.0596	0.0011	0.0075	0.0422	0.6290
355	2	0.897	0.04	-0.00	-0.0386	-0.02	15.09	0.06	15.06	0.0784	0.5891	0.0029	-0.0000	0.0004	0.7677	0.0793	0.0012	0.0095	0.0451	0.6029
355	3	0.896	1.06	-0.00	-0.0398	-0.02	15.09	0.06	15.06	0.0781	0.5873	0.0025	-0.0000	0.0004	0.9432	0.0814	0.0012	0.0095	0.0498	0.5773
355	4	0.898	3.15	-0.00	-0.0086	-0.02	15.09	0.06	15.06	0.0445	0.5856	0.0023	-0.0000	0.0004	0.9004	0.0872	0.0007	0.0103	0.0243	0.5717
355	5	0.897	6.22	-0.00	-0.1501	-0.02	15.09	0.06	15.05	0.0211	0.5768	0.0023	-0.0000	0.0002	0.5956	0.0956	0.0004	0.0110	0.0034	0.5702
355	6	0.899	9.31	-0.00	-0.2749	-0.02	15.09	0.06	15.05	0.0102	0.5863	0.0020	-0.0000	0.0002	0.6986	0.1026	0.0002	0.0115	0.0102	0.5663
355	7	0.896	12.48	-0.00	-0.3167	-0.02	15.09	0.06	15.06	0.0015	0.5776	0.0019	-0.0000	0.0002	0.7192	0.1050	0.0000	0.0121	0.0015	0.5664
355	8	0.898	0.05	-0.00	-0.0768	-0.02	15.09	0.06	15.05	0.0751	0.5996	0.0024	-0.0000	0.0004	1.0191	0.0800	0.0012	0.0095	0.0430	0.5861
356	6	0.598	0.00	0.99	28.6670	0.02	-1.85	0.06	-0.02	0.1022	0.6978	-0.0011	-0.0000	0.0002	75.5580	-0.0114	0.0002	-0.0016	-2.0967	0.2122

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key											
356	7	0.598	0.000	89.99	-2.9302	-0.02	-0.97	0.06	-0.03	0.1480	0.6175		
	6	0.001	-0.000	0.000	0.7381	7.9569	-0.0067	-0.0001	-0.0011	-1.0746	0.7710		
	9	0.0014	0.0002	0.0001	0.6220	0.6220	0.0006	-0.0001	0.0001				
356	8	0.598	-0.000	89.99	-0.1615	-0.02	-0.52	0.06	-0.02	-0.0147	0.6819		
	6	0.0014	-0.000	0.000	0.8593	0.3692	-0.0055	0.0000	-0.0007	-0.9792	0.6168		
	9	0.0016	0.0002	0.0001	0.3549	0.3549	0.0007	-0.0001	0.0000				
356	9	0.599	-0.000	89.99	-0.7989	-0.02	-0.06	0.06	-0.02	0.0044	0.8718		
	6	0.0005	-0.000	0.000	0.9315	1.9112	-0.0030	-0.0000	-0.0005	-0.8460	0.4989		
	9	0.0012	0.0002	0.0001	0.6882	0.6882	0.0009	-0.0001	0.0000				
356	10	0.599	-0.000	89.99	-0.0369	-0.01	0.46	0.06	-0.02	-0.7345	0.2405		
	6	0.0029	-0.000	-0.000	0.9895	-0.0866	-0.0019	0.0002	-0.0000	-1.0253	0.3773		
	9	0.0015	0.0003	0.0001	0.3316	0.3316	0.0008	-0.0001	0.0000				
356	11	0.599	-0.000	89.99	-1.4007	-0.02	1.01	0.06	-0.01	0.2577	-0.0395		
	6	0.0002	-0.000	0.000	0.9296	4.5892	0.0016	0.0000	-0.0000	-0.8460	0.4989		
	9	0.0010	0.0001	0.0001	0.5799	0.9120	0.0009	-0.0001	0.0000				
356	12	0.600	-0.000	89.99	-2.4507	-0.02	2.10	0.06	-0.02	0.2302	0.5577		
	6	0.0002	-0.000	0.000	0.9573	6.0275	0.0062	0.0002	0.0000	-0.9775	0.5115		
	9	0.0010	0.0002	0.0001	0.5799	0.5799	0.0007	-0.0001	0.0000				
357	1	0.600	-0.000	-90.00	1.0991	-0.02	-0.06	0.04	-1.97	-2.3046	4.1036		
	6	-0.0012	-0.000	-0.000	0.0343	0.5022	-0.0002	0.0001	-0.0001	0.4176	0.7597		
	9	-0.0003	-0.000	-0.000	0.0343	2.4940	-0.0046	-0.0003	-0.0007				
357	2	0.600	0.000	-90.00	1.0719	-0.03	-0.06	0.04	-1.01	-1.4105	0.6955		
	6	-0.0012	-0.000	-0.000	-0.0999	0.5872	-0.0006	0.0001	-0.0000	-2.4783	1.2462		
	9	-0.0005	0.0000	-0.000	0.0999	1.5075	-0.0004	-0.0002	-0.0001				
357	3	0.600	0.000	-90.00	1.2086	-0.03	-0.05	0.04	-0.52	-1.6590	1.0549		
	6	-0.0010	-0.000	-0.000	-0.1914	0.8451	-0.0005	0.0001	-0.0001	-0.4286	0.5162		
	9	-0.0005	0.0000	-0.000	0.1914	1.4343	-0.0015	-0.0001	0.0001				
357	4	0.599	0.000	-90.00	1.5899	-0.02	-0.06	0.05	-0.03	-3.4776	6.5179		
	6	-0.0007	-0.000	-0.000	-7.1743	19.2047	-0.0001	0.0000	-0.0004	0.0964	0.5738		
	9	-0.0000	0.0000	-0.000	1.0127	0.6873	0.0037	-0.0000	0.0004				
357	5	0.599	-0.000	-90.00	1.1149	-0.03	-0.06	0.04	0.46	0.6965	-3.3575		
	6	-0.0011	-0.000	-0.000	0.2277	0.6873	0.0002	0.0000	-0.0002	0.0047	0.6088		
	9	-0.0006	-0.000	-0.000	1.0127	0.6873	0.0060	0.0000	0.0000				

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Pt.	Cd	See Key											
357	6	7	0.599	0.00	-90.00	1.1464	-0.03	-0.06	0.04	0.96	-2.5712	2.2071		
	8	9	-0.0011	-0.0002	-0.0002	-5.5829	0.9513	-0.002	0.0001	-0.0001	0.0565	0.6447		
			-0.0000	0.0000	-0.0002		17.7723	0.0078	0.0000	0.0010				
357	7	7	0.599	0.00	-90.00	1.2315	-0.03	-0.06	0.04	1.97	-2.3864	0.7702		
	8	9	-0.0010	-0.0002	-0.0002	1.2877	1.0144	-0.004	0.0002	-0.0000	0.1021	0.6482		
			0.0005	0.0001	-0.0001		-1.7965	0.0128	0.0002	0.0016				
358	1	7	0.600	0.00	-0.00	0.3393	-2.02	-0.06	0.04	-0.002	10.8344	2.7933		
	8	9	-0.0057	-0.0003	-0.0006	4.3452	0.5936	0.001	0.0002	0.0000	-0.3830	0.7783		
			0.0001	0.0001	0.0000		0.1182	0.0014	-0.0001	0.0002				
358	2	7	0.599	0.00	-0.00	0.8091	-1.03	-0.05	0.04	-0.003	0.7407	0.2060		
	8	9	-0.0012	-0.0001	-0.0001	4.1429	0.7481	0.0010	0.0001	0.0000	-0.3201	0.8678		
			0.0001	0.0001	-0.0000		-1.3070	0.0015	-0.0000	0.0002				
358	3	7	0.599	0.00	-0.00	-0.0346	-0.02	-0.06	0.04	-0.002	1.6917	0.7932		
	8	9	0.0027	-0.0000	0.0003	1.7911	0.7035	0.0007	0.0002	0.0001	-0.3399	0.8212		
			0.0004	0.0001	-0.0000		-0.8571	0.0014	-0.0001	0.0002				
358	4	7	0.600	0.00	-0.00	0.0818	1.01	-0.06	0.05	-0.003	0.9324	0.3506		
	8	9	0.0072	0.0001	0.0009	1.2461	0.6723	0.0010	0.0001	0.0000	-0.4235	0.8498		
			0.0005	0.0001	-0.0000		-0.3374	0.0011	-0.0001	0.0002				
358	5	7	0.599	0.00	-0.00	0.1244	2.05	-0.06	0.04	-0.003	4.2962	2.0488		
	8	9	0.0127	0.0003	0.0015	1.0762	0.6080	0.0003	0.0002	0.0001	-0.5405	1.0151		
			0.0006	0.0001	-0.0000		-0.2655	0.0009	-0.0001	0.0001				
359	3	7	1.247	-3.18	44.99	0.0030	-3.05	-2.96	-3.00	-2.98	-0.0430	0.6875		
	8	9	-0.0345	-0.0000	-0.0047	0.0030	0.6807	-0.0348	0.0002	-0.0047	-0.0109	0.6938		
			-0.0319	0.0003	-0.0043	-0.0616	0.6786	-0.0344	0.0000	-0.0047				
359	4	7	1.248	-0.03	44.99	0.0486	-3.02	-2.94	-2.99	-2.97	-0.0524	0.7112		
	8	9	-0.0152	-0.0001	-0.0020	0.0885	0.6609	-0.0156	0.0001	-0.0022	-0.0683	0.7316		
			-0.0128	0.0002	-0.0017	-0.1718	0.6888	-0.0157	-0.0001	-0.0022	-0.1066	0.7600		
359	5	7	1.250	1.01	44.99	0.0952	-3.02	-2.94	-2.98	-2.97	-0.0683	0.7316		
	8	9	-0.0084	-0.0001	-0.0010	0.0952	0.6456	-0.0100	0.0001	-0.0014	-0.1066	0.7600		
			-0.0070	0.0002	-0.0009	-0.1718	0.6926	-0.0093	-0.0001	-0.0014				
359	6	7	1.250	3.11	44.99	0.0911	-3.00	-2.92	-2.96	-2.96	-0.2943	0.8004		
	8	9	0.0030	-0.0003	0.0004	-0.5184	0.7154	0.0004	0.0002	0.0001	2.2943	0.8102		
			0.0035	0.0003	0.0005		0.7318	0.0016	-0.0001	0.0001	-0.5322	0.8102		

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key															
359	7 7 9	1.251 0.0214 0.0229	6.26 -0.0000 0.0002	44.99 0.0028 0.0032	-0.0189 0.0627	-2.94 0.6601 0.6992	0.0190 0.0193	-2.94 0.0001 0.0002	-2.94 0.0024 0.0025	0.0521 -0.0614	0.6614 0.6451						
359	7 7 9	1.250 0.0388 0.0416	9.42 -0.0002 0.0000	44.99 0.0052 0.0056	-0.0293 0.0014	-2.92 0.6738 0.6816	0.0360 0.0394	-2.92 0.0000 0.0004	-2.92 0.0048 0.0051	0.0045 -0.0580	0.6684 0.6510						
359	7 7 9	1.250 0.0550 0.0604	12.62 -0.0004 -0.0001	45.00 0.0073 0.0081	-0.0427 -0.0135	-2.90 0.6717 0.6724	0.0530 0.0588	-2.90 -0.0002 -0.0006	-2.92 0.0071 0.0076	-0.0210 -0.0528	0.6695 0.6534						
359	10 7 9	1.250 -0.0148 -0.0130	-0.004 0.0001 0.0002	44.99 -0.0019 -0.0017	0.0547 -0.1106	-2.99 0.6701 0.6639	-2.94 -0.0152 -0.0154	-2.99 0.0001 0.0001	-2.97 -0.0022 -0.0022	-0.0417 0.0553	0.7389 0.7372						
360	1 7 9	1.249 -0.0184 -0.0159	-3.13 -0.0002 0.0002	44.99 -0.0024 -0.0021	0.0568 -0.0800	-0.06 0.6718 0.6629	-0.09 -0.0182 -0.0185	0.03 0.0001 0.0001	-0.05 -0.0025 -0.0026	-0.0364 0.0534	0.7093 0.7074						
360	2 7 9	1.250 0.0000 0.0005	-0.00 -0.0001 0.0003	44.99 0.0000 0.0001	-35.6497 3.4311	-0.04 0.0242 1.5997	-0.07 -0.010 -0.0004	0.05 0.0001 -0.0002	-0.003 -0.0001 -0.0001	-0.6231 2.1711	1.2455 1.7896						
360	3 7 9	1.250 0.0063 0.0063	1.04 -0.0001 0.0004	44.99 0.0008 0.0009	-0.1088 0.3227	-0.02 0.6628 0.7550	-0.07 0.0042 0.0051	0.07 0.0001 -0.0001	-0.005 0.0005 0.0005	0.1880 -0.1818	0.5906 0.5702						
360	4 7 9	1.250 0.0192 0.0191	3.16 -0.0001 0.0003	44.99 0.0025 0.0027	-0.0346 -0.0905	-0.00 0.6807 0.7047	-0.06 0.0159 0.0172	0.08 0.0001 -0.0002	-0.020 0.0022 -0.0022	0.0420 -0.0753	0.6531 0.6498						
360	5 7 9	1.248 0.0371 0.0400	6.30 -0.0002 0.0001	44.99 0.0050 0.0055	-0.0374 0.0134	0.01 0.6782 0.6893	-0.04 0.0343 0.0360	-0.10 0.0000 -0.0004	-0.0045 0.0047 0.0047	-0.0070 -0.0575	0.6664 0.6601						
360	6 7 9	1.250 0.0539 0.0578	9.47 -0.0005 -0.0001	44.99 0.0072 0.0078	-0.0476 -0.0087	0.03 0.6748 0.6797	-0.03 0.0512 0.0557	-0.12 0.0002 -0.0006	0.000 0.0068 0.0073	-0.0272 -0.0551	0.6676 0.6592						
360	7 7 9	1.249 0.0690 0.0761	12.65 -0.0007 -0.0002	45.00 0.0092 0.0101	-0.0537 -0.0166	0.04 0.6865 0.6635	-0.01 0.0670 0.0745	-0.14 0.0005 -0.0007	0.001 0.0088 0.0097	-0.0410 -0.0503	0.6617 0.6517						

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key														
360	8	1.246 0.0002 0.0006	0.001 0.0003	44.99 0.0001	-4.3742 3.0536	-0.02 1.4730	-0.007 -0.0003	0.06 -0.0002	-0.002 0.0001	-0.03 -0.0001	-0.6851 -2.8195	1.6874 2.4405				
361	1	1.247 -0.0130 -0.0109	-3.11 0.0002	44.99 -0.0017 -0.0014	0.0915 -0.1124	0.98 0.6639	0.98 -0.0119 -0.0132	0.99 0.0000 -0.0002	0.94 -0.0019	-0.0307 0.0972	0.7379 0.7186					
361	2	1.249 0.0051 0.0050	0.02 0.0001 0.0004	44.99 0.0007 0.0008	-0.1542 0.4251	1.01 0.6810 0.8502	1.01 0.0044 0.0045	1.02 0.0001 -0.0001	0.95 0.0004	0.1618 -0.2172	0.5888 0.5268					
361	3	1.249 0.0114 0.0112	1.06 0.0001 0.0004	44.99 0.0015 0.0016	-0.0645 0.1817	1.00 0.6693 0.7324	1.02 0.0099 0.0103	1.02 0.0001 -0.0002	0.96 0.0012	0.0618 -0.1051	0.6488 0.6062					
361	4	1.248 0.0247 0.0246	3.15 0.0001 0.0003	44.99 0.0033 0.0034	-0.0371 0.0618	1.02 0.6707 0.7047	1.03 0.0215 0.0227	1.04 0.0000 -0.0003	0.96 0.0028 0.0029	0.0175 -0.0686	0.6665 0.6572					
361	5	1.247 0.0424 0.0451	6.31 0.0003 0.0000	44.99 0.0057 0.0062	-0.0447 0.0092	1.04 0.6757 0.6906	1.04 0.0401 0.0415	1.06 -0.0001 -0.0004	0.98 0.0053 0.0055	-0.0208 -0.0581	0.6665 0.6640					
361	6	1.249 0.0590 0.0631	9.47 0.0006 0.0001	44.99 0.0079 0.0085	-0.0527 -0.0108	1.06 0.6717 0.6743	1.06 0.0568 0.0612	1.08 -0.0003 -0.0006	0.99 0.0075 0.0080	-0.0346 -0.0542	0.6656 0.6579					
361	7	1.248 0.0733 0.0803	12.67 0.0008 0.0002	45.00 0.0097 0.0106	-0.0563 -0.0154	1.07 0.6845 0.6617	1.07 0.0721 0.0796	1.10 -0.0006 -0.0008	0.99 0.0095 0.0103	-0.0451 -0.0502	0.6610 0.6502					
361	8	1.248 0.0052 0.0051	0.01 0.0001 0.0004	44.99 0.0006 0.0008	-0.1644 0.4178	1.00 0.6503 0.7838	1.01 0.0046 0.0044	1.02 0.0001 -0.0001	0.95 0.0005 0.0004	0.1285 -0.2113	0.5345 0.5297					
362	1	1.248 -0.0026 -0.0025	-3.09 0.0002 0.0003	44.99 -0.0002 -0.0002	0.4460 -0.6192	3.01 0.5433 0.4428	2.97 -0.0019 -0.0024	2.91 0.0000 -0.0003	3.02 -0.0003 -0.0004	-0.1931 -0.6258	0.9347 0.8767					
362	2	1.250 0.0156 0.0143	0.03 0.0001 0.0003	44.99 0.0021 0.0021	-0.0537 -0.1324	3.03 0.6804 0.7467	2.99 0.0152 0.0150	2.95 0.0000 -0.0002	3.05 0.0019 0.0019	0.0269 -0.0681	0.6522 0.6613					

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt. Cd \leftarrow See Key \rightarrow

362	3	7	5	1.248	-0.0002	1.07	0.0003	44.99	0.0030	-0.0489	0.0776	3.04	0.6800	0.7201	3.00	0.0209	0.0214	-0.0002	2.95	0.0000	0.0028	3.05	0.0027	0.0076	0.6631	0.6686
362	4	7	9	1.248	-0.0003	3.19	0.0001	44.99	0.0048	-0.0424	0.0271	3.06	0.6771	0.7029	3.01	0.0324	0.0344	-0.0004	2.97	0.0000	0.0043	3.06	0.0043	-0.0108	0.6677	0.6676
362	5	7	9	1.250	-0.0005	6.33	0.0000	44.99	0.0071	-0.0533	0.0037	3.07	0.6755	0.6827	3.03	0.0507	0.0530	-0.0006	2.99	0.0003	0.0067	3.07	0.0067	-0.0349	0.6686	0.6674
362	6	7	9	1.248	-0.0001	9.51	0.0008	44.99	0.0091	-0.0582	0.0136	3.09	0.6676	0.6683	3.04	0.0667	0.0722	-0.0007	3.01	0.0005	0.0088	3.08	0.0088	-0.0442	0.6628	0.6547
362	7	7	9	1.248	-0.0010	12.68	0.0002	44.99	0.0108	-0.0610	0.0166	3.10	0.6571	0.6531	3.05	0.0813	0.0902	-0.0009	3.03	0.0008	0.0106	3.09	0.0106	-0.0504	0.6540	0.6438
362	8	7	9	1.249	-0.0001	0.02	0.0003	44.99	0.0021	-0.0544	0.1352	3.04	0.6790	0.7419	2.99	0.0154	0.0151	-0.0002	2.95	0.0000	0.0020	3.05	0.0019	0.0233	0.6452	0.6604
363	1	7	9	1.249	-0.0001	3.05	0.0003	44.99	0.0016	-0.0812	0.1492	6.02	0.6860	0.7501	6.02	0.0133	0.0117	-0.0002	6.06	0.0000	0.0018	6.00	0.0018	0.0350	0.6857	0.6769
363	2	7	9	1.248	-0.0003	0.07	0.0001	44.99	0.0043	-0.0500	0.0256	6.04	0.6899	0.7139	6.04	0.0314	0.0313	-0.0003	6.08	0.0000	0.0042	6.02	0.0042	-0.0116	0.6699	0.6803
363	3	7	9	1.248	-0.0003	1.12	0.0000	44.99	0.0052	-0.0515	0.0131	6.05	0.6896	0.7079	6.04	0.0371	0.0379	-0.0004	6.09	0.0001	0.0051	6.02	0.0049	-0.0209	0.6734	0.6808
363	4	7	9	1.249	-0.0005	3.22	0.0000	44.99	0.0069	-0.0517	0.0050	6.06	0.6763	0.6882	6.05	0.0485	0.0508	-0.0006	6.11	0.0003	0.0065	6.03	0.0065	-0.0346	0.6720	0.6718
363	5	7	9	1.249	-0.0001	6.35	0.0008	44.99	0.0090	-0.0594	0.0138	6.08	0.6697	0.6693	6.07	0.0661	0.0688	-0.0007	6.13	0.0006	0.0087	6.04	0.0087	-0.0463	0.6650	0.6629

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key										
363	6	7	1.248 0.0821 0.0873	9.54 -0.0010 -0.0002	44.99 0.0108 0.0114	-0.0631 -0.0170	6.09 0.6587 0.6556	6.07 0.0810 0.0874	-0.0008 -0.0009	6.05 0.0106 0.0113	-0.0511 -0.0546	0.6540 0.6474
363	7	6	1.248 0.0940 0.1031	12.73 -0.0011 -0.0004	44.99 0.0121 0.0131	-0.0607 -0.0219	6.10 0.6460 0.6364	6.09 0.0938 0.1042	6.16 -0.0010 -0.0011	6.06 0.0120 0.0131	-0.0535 -0.0531	0.6421 0.6303
364	3	7	1.249 0.0281 0.0278	-3.04 -0.0002 -0.0000	44.99 0.0041 0.0039	-0.0460 -0.0054	9.05 0.7324 0.7051	9.08 0.0284 0.0274	9.02 0.0001 -0.0004	8.98 0.0041 0.0039	0.0180 -0.0753	0.7236 0.7143
364	4	7	1.251 0.0471 0.0476	0.06 -0.0004 -0.0002	44.99 0.0066 0.0066	-0.0508 -0.0272	9.06 0.7066 0.6944	9.09 0.0464 0.0475	9.03 -0.0001 -0.0005	9.00 0.0064 0.0065	-0.0179 -0.0585	0.6940 0.6924
364	5	7	1.249 0.0538 0.0542	1.14 -0.0005 -0.0002	44.99 0.0075 0.0074	-0.0517 -0.0275	9.07 0.6980 0.6887	9.10 0.0521 0.0544	9.04 -0.0002 -0.0006	9.00 0.0071 0.0074	-0.0258 -0.0570	0.6879 0.6823
364	6	7	1.249 0.0661 0.0677	3.24 -0.0007 -0.0003	44.99 0.0090 0.0091	-0.0542 -0.0281	9.08 0.6819 0.6746	9.11 0.0633 0.0666	9.06 -0.0004 -0.0007	9.01 0.0086 0.0090	-0.0364 -0.0581	0.6808 0.6759
364	7	8	1.249 0.0815 0.0860	6.37 -0.0009 -0.0004	44.99 0.0109 0.0113	-0.0591 -0.0253	9.10 0.6691 0.6585	9.12 0.0798 0.0840	9.08 -0.0006 -0.0009	9.02 0.0106 0.0110	-0.0427 -0.0562	0.6646 0.6569
364	8	7	1.250 0.0943 0.1008	9.56 -0.0011 -0.0005	44.99 0.0123 0.0129	-0.0622 -0.0279	9.11 0.6526 0.6394	9.14 0.0929 0.1012	9.10 -0.0008 -0.0010	9.04 0.0121 0.0128	-0.0460 -0.0538	0.6510 0.6370
364	9	7	1.249 0.1044 0.1158	12.72 -0.0012 -0.0006	44.99 0.0134 0.0144	-0.0591 -0.0273	9.12 0.6435 0.6234	9.14 0.1044 0.1168	9.12 -0.0010 -0.0011	9.05 0.0133 0.0145	-0.0520 -0.0501	0.6400 0.6213
364	10	7	1.248 0.0474 0.0474	0.08 -0.0004 -0.0001	44.99 0.0066 0.0066	-0.0516 -0.0207	9.07 0.7012 0.6992	9.09 0.0470 0.0477	9.04 -0.0002 -0.0005	9.01 0.0064 0.0065	-0.0212 -0.0588	0.6855 0.6829
365	1	7	1.250 0.0605 0.0593	-2.98 -0.0006 -0.0004	44.99 0.0083 0.0081	-0.0548 -0.0378	15.07 0.6902 0.6891	15.05 0.0606 0.0602	15.06 -0.0003 -0.0009	15.02 0.0083 0.0081	-0.0298 -0.0760	0.6884 0.6745

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																										
365	2	7	1.249	0.0776	0.0780	0.16	-0.0008	-0.0005	44.99	0.0104	0.0104	-0.0547	-0.0336	15.09	0.6738	0.6705	15.07	0.0769	0.0787	15.08	-0.0005	-0.0009	15.04	0.0103	0.0103	-0.0370	-0.0614	0.6700	0.6589
365	3	7	1.248	0.0834	0.0837	1.17	-0.0009	-0.0005	44.99	0.0111	0.0111	-0.0554	-0.0312	15.09	0.6663	0.6632	15.07	0.0816	0.0848	15.09	-0.0006	-0.0010	15.03	0.0108	0.0110	-0.0369	-0.0608	0.6659	0.6505
365	4	7	1.248	0.0937	0.0948	3.30	-0.0010	-0.0005	44.99	0.0122	0.0122	-0.0566	-0.0299	15.10	0.6539	0.6476	15.08	0.0908	0.0958	15.10	-0.0007	-0.0011	15.04	0.0119	0.0122	-0.0413	-0.0612	0.6563	0.6390
365	5	7	1.247	0.1052	0.1105	6.42	-0.0012	-0.0006	44.99	0.0134	0.0139	-0.0586	-0.0306	15.11	0.6380	0.6298	15.09	0.1036	0.1111	15.12	-0.0009	-0.0012	15.05	0.0133	0.0138	-0.0445	-0.0579	0.6420	0.6222
365	6	7	1.247	0.1151	0.1222	9.59	-0.0015	-0.0007	45.00	0.0144	0.0149	-0.0662	-0.0293	15.12	0.6295	0.6132	15.09	0.1150	0.1249	15.13	-0.0012	-0.0013	15.06	0.0145	0.0151	-0.0562	-0.0525	0.6333	0.6059
365	7	7	1.249	0.1226	0.1309	12.75	-0.0016	-0.0005	45.00	0.0153	0.0158	-0.0663	-0.0228	15.12	0.6271	0.6041	15.10	0.1219	0.1312	15.15	-0.0014	-0.0010	15.08	0.0153	0.0155	-0.0599	-0.0389	0.6283	0.5943
365	8	7	1.249	0.0780	0.0781	0.15	-0.0008	-0.0004	44.99	0.0104	0.0104	-0.0552	-0.0315	15.08	0.6705	0.6708	15.07	0.0773	0.0790	15.09	-0.0005	-0.0009	15.03	0.0103	0.0103	-0.0380	-0.0617	0.6661	0.6562
366	1	7	0.898	0.0696	0.0657	-2.97	0.0006	0.0011	44.99	0.0087	0.0082	0.0439	0.0857	15.08	0.6292	0.6270	15.07	0.0691	0.0693	15.07	0.0010	0.0005	15.05	0.0086	0.0085	0.0744	0.0391	0.6265	0.6142
366	2	7	0.899	0.0842	0.0792	0.11	0.0007	0.0012	44.99	0.0102	0.0094	0.0418	0.0812	15.10	0.6063	0.5945	15.08	0.0842	0.0823	15.10	0.0009	0.0007	15.06	0.0103	0.0097	0.0575	0.0455	0.6117	0.5891
366	3	7	0.898	0.0884	0.0818	1.15	0.0007	0.0013	44.99	0.0106	0.0096	0.0412	0.0831	15.10	0.6020	0.5897	15.08	0.0880	0.0854	15.09	0.0009	0.0008	15.06	0.0106	0.0099	0.0525	0.0494	0.6050	0.5804
366	4	7	0.900	0.0949	0.0872	3.19	0.0006	0.0010	44.99	0.0111	0.0101	0.0365	0.0590	15.10	0.5860	0.5820	15.09	0.0950	0.0907	15.09	0.0008	0.0007	15.06	0.0113	0.0104	0.0450	0.0393	0.5954	0.5757

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TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd																			
366	5	7	0.900	6.29	44.99	0.0118	0.0140	15.11	15.09	15.09	15.09	15.09	15.09	15.06	0.0293	0.5854				
			0.1016	0.0002	0.0118	0.0109	0.0293	0.5806	0.1021	0.0006	0.0002	0.0006	0.0119	0.0119	0.5708					
			0.0942	0.0005	0.0109	0.0109	0.0293	0.5786	0.0971	0.0002	0.0002	0.0110	0.0110	0.5854						
366	6	7	0.900	9.40	44.99	0.0120	-0.0009	15.10	15.09	15.09	15.09	15.06	0.0080	0.5829						
			0.1039	-0.0000	0.0120	0.0117	0.0179	0.5812	0.1055	0.0001	0.0001	0.0123	0.0087	0.5689						
			0.1013	0.0003	0.0117	0.0119	0.0179	0.5817	0.1020	0.0001	0.0001	0.0116	0.0087	0.5689						
366	7	7	0.899	12.51	45.00	0.0124	-0.0121	15.10	15.08	15.10	15.10	15.06	-0.0051	0.5809						
			0.1071	-0.0002	0.0124	0.0119	0.0179	0.5804	0.1124	-0.0001	0.0000	0.0130	-0.0051	0.5651						
			0.1036	0.0003	0.0119	0.0119	0.0179	0.5749	0.1064	0.0000	0.0000	0.0120	0.0013	0.5651						
366	8	7	0.897	0.11	44.99	0.0102	0.0417	15.09	15.08	15.10	15.10	15.06	0.0580	0.6096						
			0.0843	0.0007	0.0102	0.0094	0.0842	0.6068	0.0842	0.0009	0.0007	0.0102	0.0461	0.5902						
			0.0792	0.0013	0.0094	0.0094	0.0842	0.5991	0.0822	0.0007	0.0007	0.0097	0.0461	0.5902						
367	1	7	0.899	-3.02	44.99	0.0056	0.0211	9.05	9.09	9.02	9.02	9.01	0.0803	0.6864						
			0.0391	0.0001	0.0056	0.0047	0.1048	0.7175	0.0378	0.0006	0.0001	0.0052	0.0259	0.6864						
			0.0349	0.0007	0.0047	0.0047	0.1048	0.6824	0.0374	0.0001	0.0001	0.0050	0.0259	0.6773						
367	2	7	0.898	0.07	44.99	0.0072	0.0589	9.08	9.11	9.05	9.05	9.03	0.0862	0.6373						
			0.0557	0.0006	0.0072	0.0067	0.1203	0.6505	0.0544	0.0009	0.0007	0.0069	0.0626	0.6408						
			0.0529	0.0012	0.0067	0.0067	0.1203	0.6376	0.0559	0.0007	0.0007	0.0071	0.0626	0.6408						
367	3	7	0.897	1.12	44.99	0.0078	0.0582	9.09	9.12	9.06	9.06	9.02	0.0833	0.6292						
			0.0617	0.0007	0.0078	0.0074	0.1153	0.6375	0.0599	0.0009	0.0008	0.0075	0.0626	0.6324						
			0.0593	0.0013	0.0074	0.0074	0.1153	0.6298	0.0613	0.0008	0.0008	0.0077	0.0626	0.6324						
367	4	7	0.898	3.17	44.99	0.0090	0.0493	9.09	9.13	9.07	9.07	9.04	0.0758	0.6199						
			0.0730	0.0007	0.0090	0.0086	0.1022	0.6206	0.0705	0.0010	0.0009	0.0087	0.0671	0.6102						
			0.0710	0.0014	0.0086	0.0086	0.1022	0.6091	0.0710	0.0009	0.0009	0.0086	0.0671	0.6102						
367	5	7	0.898	6.24	44.99	0.0102	0.0484	9.11	9.14	9.08	9.08	9.04	0.0614	0.5998						
			0.0860	0.0008	0.0102	0.0092	0.0885	0.5957	0.0848	0.0010	0.0010	0.0101	0.0635	0.5998						
			0.0801	0.0014	0.0092	0.0092	0.0885	0.5790	0.0825	0.0010	0.0010	0.0096	0.0635	0.5998						
367	6	7	0.898	9.39	44.99	0.0109	0.0334	9.11	9.15	9.07	9.07	9.04	0.0500	0.5819						
			0.0933	0.0006	0.0109	0.0100	0.0446	0.5858	0.0926	0.0009	0.0005	0.0107	0.0325	0.5819						
			0.0869	0.0007	0.0100	0.0100	0.0446	0.5785	0.0904	0.0005	0.0005	0.0103	0.0325	0.5819						
367	7	7	0.898	12.50	45.00	0.0001	0.0098	9.11	9.14	9.08	9.08	9.04	0.0220	0.5813						
			0.0998	0.0001	0.0001	0.0004	0.0250	0.5813	0.0995	0.0004	0.0003	0.0115	0.0172	0.5813						
			0.0940	0.0004	0.0004	0.0109	0.0250	0.5808	0.0939	0.0003	0.0003	0.0107	0.0172	0.5813						

See Key

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key													
367	8	0.898 0.0553 0.0530	0.0006 0.0013	0.072 0.0067	0.0614 0.1228	9.08 0.6504 0.6369	9.11 0.0543 0.0561	9.05 0.0009 0.0006	9.02 0.0069 0.0071	0.0867 0.0616	0.6351 0.6358				
368	1	0.898 0.0155 0.0148	-3.06 0.0002 0.0006	4.99 0.0022 0.0020	0.0701 0.2056	6.03 0.7295 0.6966	6.02 0.0136 0.0165	6.06 0.0006 0.0000	6.01 0.0020 0.0022	0.2514 0.0180	0.7367 0.6829				
368	2	0.899 0.0423 0.0373	0.005 0.0001 0.0009	4.99 0.0059 0.0045	0.0228 0.1268	6.05 0.6975 0.6669	6.04 0.0388 0.0404	6.08 0.0005 0.0003	6.03 0.0052 0.0054	0.0768 0.0442	0.6775 0.6703				
368	3	0.900 0.0478 0.0431	1.08 0.0004 0.0012	4.99 0.0065 0.0056	0.0497 0.1390	6.06 0.6813 0.6516	6.04 0.0439 0.0464	6.09 0.0007 0.0005	6.04 0.0058 0.0062	0.0861 0.0597	0.6606 0.6685				
368	4	0.898 0.0576 0.0565	3.16 0.0008 0.0014	4.99 0.0073 0.0070	0.0703 0.1258	6.08 0.6403 0.6230	6.06 0.0537 0.0566	6.11 0.0010 0.0008	6.05 0.0068 0.0072	0.0997 0.0768	0.6355 0.6373				
368	5	0.898 0.0749 0.0728	6.26 0.0008 0.0014	4.99 0.0091 0.0086	0.0556 0.1022	6.10 0.5968 0.5782	6.08 0.0716 0.0714	6.13 0.0011 0.0009	6.06 0.0088 0.0085	0.0814 0.0698	0.6165 0.6011				
368	6	0.899 0.0855 0.0809	9.37 0.0008 0.0013	45.00 0.0102 0.0093	0.0523 0.0808	6.11 0.5961 0.5782	6.08 0.0843 0.0832	6.13 0.0010 0.0010	6.07 0.0100 0.0095	0.0628 0.0638	0.5978 0.5741				
368	7	0.898 0.0933 0.0868	12.49 0.0005 0.0007	45.00 0.0108 0.0100	0.0321 0.0419	6.10 0.5829 0.5772	6.09 0.0918 0.0890	6.13 0.0009 0.0004	6.06 0.0106 0.0101	0.0504 0.0266	0.5823 0.5703				
368	8	0.899 0.0423 0.0376	0.005 0.0001 0.0009	4.99 0.0059 0.0050	0.0215 0.1254	6.05 0.6993 0.6692	6.03 0.0391 0.0405	6.09 0.0005 0.0003	6.02 0.0053 0.0054	0.0710 0.0434	0.6786 0.6718				
369	1	0.899 -0.0010 -0.0020	-3.07 0.0002 0.0001	4.99 0.0000 0.0002	1.1836 -0.4681	3.01 -0.0898 0.5300	2.97 -0.0036 -0.0002	2.92 0.0001 -0.0003	3.03 -0.0004 -0.0000	-0.2216 -9.3531	0.6706 2.1466				
369	2	0.897 0.0211 0.0179	0.001 0.0007	4.99 0.0030 0.0024	0.0285 0.2005	3.04 0.7115 0.6932	2.99 0.0164 0.0205	2.95 0.0006 0.0000	3.04 0.0022 0.0027	0.1862 0.0229	0.6898 0.6606				

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key →																		
369	3	0.901	1.06	44.99	0.0192	3.04	3.00	2.96	3.05	0.1213	0.6025	0.0302	0.0001	0.0042	0.7047	0.0242	0.0005	0.0033	0.0381	0.6745
		0.0263	0.0008	0.0036	0.1519	0.6921	0.0275	0.0002	0.0037	0.0381	0.6025	0.0302	0.0001	0.0042	0.7047	0.0242	0.0005	0.0033	0.0381	0.6745
369	4	0.901	5.12	44.99	0.0493	3.06	3.01	2.98	3.06	0.0912	0.6743	0.0448	0.0004	0.0061	3.853	0.0384	0.0007	0.0051	0.0912	0.6743
		0.0407	0.0011	0.0053	0.1394	0.6601	0.0420	0.0004	0.0056	0.0558	0.6743	0.0448	0.0004	0.0056	3.853	0.0384	0.0007	0.0056	0.0558	0.6743

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																							
369	5	7	0.899	6.23	45.00	0.0724	0.1258	3.09	3.04	3.00	3.09	0.6324	0.6219	0.553	0.0579	3.00	0.0011	0.0009	3.02	3.10	0.0840	0.0730	0.1036	0.0837	0.6301	0.6315
		6	0.0606	0.0008	0.0076			0.6324	0.0553	0.0011	0.0069					0.0009			0.0012	0.0088						
		9	0.0595	0.0014	0.0074			0.6219	0.0579	0.0009	0.0073					0.0009			0.0010	0.0087						
369	6	7	0.898	9.36	45.00	0.0622	0.0987	3.10	3.05	3.02	3.10	0.6130	0.5955	0.723	0.0740	3.02	0.0012	0.0010	3.02	3.10	0.0840	0.0730	0.1036	0.0837	0.6135	0.5915
		6	0.0752	0.0009	0.0092			0.6130	0.0723	0.0012	0.0088					0.0012			0.0010	0.0087						
		9	0.0740	0.0014	0.0088			0.5955	0.0740	0.0010	0.0087					0.0010			0.0010	0.0087						
369	7	7	0.899	12.48	45.00	0.0573	0.0766	3.11	3.05	3.02	3.11	0.5995	0.5803	0.850	0.0847	3.02	0.0010	0.0009	3.02	3.11	0.0617	0.0541	0.0617	0.0541	0.5962	0.5726
		6	0.0860	0.0009	0.0103			0.5995	0.0850	0.0010	0.0097					0.0010			0.0009	0.0097						
		9	0.0818	0.0012	0.0094			0.5803	0.0847	0.0009	0.0097					0.0009			0.0009	0.0097						
369	8	7	0.898	0.00	44.99	0.0298	0.2094	3.04	2.99	2.95	3.05	0.7122	0.6943	0.162	0.0204	2.95	0.0005	0.0000	2.95	3.05	0.1817	0.0211	0.1817	0.0211	0.6808	0.6545
		6	0.0210	0.0001	0.0029			0.7122	0.0162	0.0005	0.0022					0.0005			0.0000	0.0022						
		9	0.0177	0.0007	0.0024			0.6943	0.0204	0.0000	0.0026					0.0000			0.0000	0.0026						
370	1	7	0.899	-3.09	44.99	0.2190	0.0569	0.97	0.98	0.98	0.93	0.6264	0.6754	-0.0150	-0.0132	0.98	-0.0001	-0.0006	0.98	0.93	0.0578	0.2431	0.0578	0.2431	0.6891	0.7041
		6	-0.0134	-0.0005	-0.0016			0.6264	-0.0150	-0.0001	-0.0020					-0.0001			-0.0006	-0.0020						
		9	-0.0116	-0.0001	-0.0015			0.6754	-0.0132	-0.0006	-0.0018					-0.0006			-0.0006	-0.0018						
370	2	7	0.898	0.00	44.99	0.1154	0.4495	1.00	1.01	1.02	0.95	0.7971	0.7378	0.022	0.0061	1.02	0.0004	0.0000	1.02	0.95	0.9473	0.0584	0.9473	0.0584	0.6053	0.5956
		6	0.0051	0.0001	0.0008			0.7971	0.0022	0.0004	0.0002					0.0004			0.0004	0.0002						
		9	0.0058	0.0005	0.0008			0.7378	0.0061	0.0000	0.0007					0.0000			0.0000	0.0007						
370	3	7	0.897	1.03	44.99	0.0667	0.2806	1.01	1.02	1.03	0.96	0.7271	0.7098	0.087	0.0132	1.03	0.0005	0.0000	1.03	0.96	0.3310	0.0257	0.3310	0.0257	0.6826	0.6427
		6	0.0136	0.0001	0.0019			0.7271	0.0087	0.0005	0.0012					0.0005			0.0005	0.0012						
		9	0.0127	0.0007	0.0018			0.7098	0.0132	0.0000	0.0017					0.0000			0.0000	0.0017						
370	4	7	0.900	3.09	44.99	0.0499	0.1378	1.03	1.03	1.05	0.97	0.7007	0.6846	0.241	0.0283	1.05	0.0007	0.0002	1.05	0.97	0.1463	0.0460	0.1463	0.0460	0.6796	0.6770
		6	0.0311	0.0003	0.0043			0.7007	0.0241	0.0007	0.0032					0.0007			0.0007	0.0032						
		9	0.0305	0.0008	0.0041			0.6846	0.0283	0.0002	0.0038					0.0002			0.0002	0.0038						
370	5	7	0.897	6.21	45.00	0.0801	0.1376	1.05	1.06	1.08	0.99	0.6534	0.6276	0.0450	0.0493	1.08	0.0010	0.0007	1.08	0.99	0.1144	0.0747	0.1144	0.0747	0.6462	0.6590
		6	0.0503	0.0008	0.0065			0.6534	0.0450	0.0010	0.0058					0.0010			0.0010	0.0058						
		9	0.0505	0.0013	0.0063			0.6276	0.0493	0.0007	0.0065					0.0007			0.0007	0.0065						
370	6	7	0.898	9.32	45.00	0.0819	0.1089	1.07	1.08	1.09	1.01	0.6291	0.6041	0.0614	0.0657	1.09	0.0013	0.0010	1.09	1.01	0.1069	0.0791	0.1069	0.0791	0.6216	0.6034
		6	0.0646	0.0010	0.0081			0.6291	0.0614	0.0013	0.0076					0.0013			0.0013	0.0076						
		9	0.0670	0.0014	0.0080			0.6041	0.0657	0.0010	0.0079					0.0010			0.0010	0.0079						
370	7	7	0.897	12.46	45.00	0.0710	0.0909	1.09	1.09	1.10	1.02	0.6117	0.5837	0.0766	0.0796	1.10	0.0013	0.0011	1.10	1.02	0.0860	0.0690	0.0860	0.0690	0.6077	0.5836
		6	0.0782	0.0011	0.0092			0.6117	0.0766	0.0013	0.0093					0.0013			0.0013	0.0093						
		9	0.0792	0.0014	0.0092			0.5837	0.0796	0.0011	0.0093					0.0011			0.0011	0.0093						

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key															
370	8	7 6 9	0.899 0.0051 0.0059	-0.001 0.0005	44.99 0.0008	0.1163 0.4479	1.00 0.8157 0.7189	1.01 0.0024 0.0001	1.01 0.0004 -0.0000	0.96 0.0002 0.0007	0.8541 -0.0365	0.5746 0.6164						
371	1	7 6 9	0.898 -0.0206 -0.0193	-3.11 -0.0007 -0.0001	44.99 -0.0026 -0.0025	0.1837 0.0272	-0.06 0.6503 0.6607	-0.10 -0.0226 -0.0209	0.03 -0.0002 -0.0007	-0.04 -0.0031 -0.0029	0.0624 0.1677	0.6946 0.7057						
371	2	7 6 9	0.900 0.0008 0.0008	-0.00 -0.0001 0.0003	44.99 0.0002 0.0001	-0.8743 1.9390	-0.03 1.4633 0.6726	-0.07 -0.0022 0.0017	0.06 0.0001 -0.0003	-0.02 -0.0001 -0.0001	-0.4032 -0.8463	0.8199 0.5249						
371	3	7 6 9	0.900 0.0058 0.0071	1.01 0.0001 0.0005	44.99 0.0009 0.0009	0.1298 0.3995	-0.02 0.7919 0.7023	-0.06 0.0021 0.0064	0.07 0.0004 -0.0000	-0.01 0.0002 0.0007	1.0183 -0.0010	0.6371 0.6163						
371	4	7 6 9	0.899 0.0238 0.0236	3.08 0.0002 0.0008	44.99 0.0033 0.0032	0.0581 0.1708	-0.00 0.6943 0.6894	-0.05 0.0163 0.0218	0.09 0.0006 0.0001	-0.00 0.0021 0.0028	0.2068 0.0285	0.6711 0.6589						
371	5	7 6 9	0.898 0.0463 0.0457	6.21 0.0006 0.0012	44.99 0.0061 0.0058	0.0668 0.1383	0.01 0.6655 0.6413	-0.01 0.0400 0.0438	0.12 0.0008 0.0005	0.02 0.0053 0.0059	0.1100 0.0648	0.6631 0.6740						
371	6	7 6 9	0.899 0.0592 0.0629	9.33 0.0010 0.0014	45.00 0.0075 0.0076	0.0918 0.1127	0.03 0.6394 0.6091	0.00 0.0559 0.0618	0.14 0.0013 0.0009	0.03 0.0070 0.0076	0.1178 0.0800	0.6285 0.6185						
371	7	7 6 9	0.900 0.0736 0.0770	12.46 0.0011 0.0013	45.00 0.0090 0.0090	0.0774 0.0890	0.05 0.6150 0.5881	0.02 0.0719 0.0771	0.15 0.0013 0.0010	0.04 0.0088 0.0090	0.0937 0.0662	0.6145 0.5875						
371	8	7 6 9	0.900 0.0007 0.0006	-0.02 -0.0001 0.0003	44.99 0.0002 0.0000	-1.0031 2.7095	-0.03 1.6597 0.6205	-0.06 -0.0023 0.0018	0.05 0.0001 -0.0003	-0.02 -0.0003 -0.0001	-0.4212 -0.8162	0.7960 0.4469						
372	1	7 6 9	0.898 -0.0392 -0.0400	-3.15 -0.0011 -0.0001	44.99 -0.0051 -0.0055	0.1480 0.0227	-3.05 0.6565 0.6890	-2.99 -0.0422 -0.0410	-3.03 -0.0005 -0.0008	-3.00 -0.0057 -0.0054	0.0703 0.1004	0.6792 0.6673						
372	2	7 6 9	0.899 -0.0162 -0.0164	-0.04 -0.0006 -0.0000	44.99 -0.0020 -0.0023	0.2103 0.0166	-3.02 0.6454 0.7041	-2.96 -0.0197 -0.0183	-3.00 -0.0002 -0.0005	-2.98 -0.0027 -0.0026	0.0519 0.1602	0.6879 0.7207						

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																		
372	3	7	0.200	0.0084	0.0078	0.0004	0.0000	0.0000	0.0004	0.0010	0.2628	0.6371	-2.93	-0.0114	-0.0092	-2.99	-0.0001	-0.0016	0.0588	0.7208	
		5	-0.0078	-0.0000	-0.0011	44.99	-0.0011	0.0320	0.7124	-0.0092	-0.0005	-2.99	-0.0001	-0.0016	0.0588	0.7208					
372	4	7	0.899	0.0028	0.0042	3.06	0.0004	0.0005	0.2030	0.5496	-2.99	-0.0012	-0.0003	-0.0002	-2.96	-0.0002	-0.0003	-1.2497	0.8201	0.5675	
		5	0.0042	0.0004	0.0005	44.99	0.0004	0.0005	0.2030	0.5496	-2.99	-0.0012	-0.0003	-0.0002	-2.96	-0.0002	-0.0003	-1.2497	0.8201	0.5675	
372	5	7	0.900	0.0268	0.0292	6.18	0.0007	0.0039	0.0792	0.1276	-2.96	-0.0115	-0.0040	-0.0001	-2.93	-0.0001	-0.0001	0.2144	0.6728	0.6476	
		5	0.0292	0.0007	0.0039	45.00	0.0039	0.1276	0.0792	0.1276	-2.96	-0.0115	-0.0040	-0.0001	-2.93	-0.0001	-0.0001	0.2144	0.6728	0.6476	
372	6	7	0.901	0.0438	0.0482	9.32	0.0010	0.0061	0.1141	0.1284	-2.94	-0.0399	-0.0493	-0.0005	-2.91	-0.0011	-0.0052	0.1478	0.6539	0.6614	
		5	0.0482	0.0010	0.0061	45.00	0.0061	0.1284	0.1141	0.1284	-2.94	-0.0399	-0.0493	-0.0005	-2.91	-0.0011	-0.0052	0.1478	0.6539	0.6614	
372	7	7	0.900	0.0587	0.0659	12.44	0.0012	0.0079	0.1084	0.0998	-2.92	-0.0558	-0.0672	-0.0008	-2.90	-0.0015	-0.0082	0.1376	0.6310	0.6113	
		5	0.0659	0.0012	0.0079	45.00	0.0079	0.0998	0.1084	0.0998	-2.92	-0.0558	-0.0672	-0.0008	-2.90	-0.0015	-0.0082	0.1376	0.6310	0.6113	
372	8	7	0.898	0.0162	0.0162	-0.0006	-0.0000	-0.0023	0.2049	0.0115	-3.02	-0.0380	-0.0179	-0.0002	-3.00	-0.0002	-0.0026	0.0639	0.7105	0.7298	
		5	0.0162	-0.0006	-0.0023	44.99	-0.0023	0.0115	0.2049	0.0115	-3.02	-0.0380	-0.0179	-0.0002	-3.00	-0.0002	-0.0026	0.0639	0.7105	0.7298	
374	3	7	1.249	0.0042	0.0031	-3.23	0.0003	0.0004	0.3736	0.0230	-0.01	-0.0326	-0.0908	-0.0017	-0.05	-0.0019	-0.0173	0.1045	0.9570	0.9681	
		5	0.0031	-0.0003	-0.0004	-0.0004	0.0004	0.0230	0.3736	0.0230	-0.01	-0.0326	-0.0908	-0.0017	-0.05	-0.0019	-0.0173	0.1045	0.9570	0.9681	
374	4	7	1.251	0.0065	0.0040	-0.0003	0.0000	0.0006	0.2751	0.0117	-0.00	-0.0300	-0.0306	-0.0009	-0.06	-0.0009	-0.0057	0.1588	0.9559	1.0087	
		5	0.0040	-0.0003	-0.0006	-0.0006	0.0006	0.0117	0.2751	0.0117	-0.00	-0.0300	-0.0306	-0.0009	-0.06	-0.0009	-0.0057	0.1588	0.9559	1.0087	
374	5	7	1.251	0.0066	0.0041	0.99	0.0003	0.0007	0.2649	0.0407	-0.00	-0.0357	-0.0112	-0.0003	-0.06	-0.0003	-0.0025	0.2106	1.0225	1.1170	
		5	0.0041	0.0003	0.0007	-0.0007	0.0007	0.0407	0.2649	0.0407	-0.00	-0.0357	-0.0112	-0.0003	-0.06	-0.0003	-0.0025	0.2106	1.0225	1.1170	
374	6	7	1.251	0.0052	0.0042	3.14	0.0001	0.0006	0.2668	0.1190	-0.01	-0.0356	-0.0219	-0.0004	-0.06	-0.0004	-0.0038	0.1114	0.9088	0.8819	
		5	0.0042	0.0001	0.0006	-0.0006	0.0006	0.1190	0.2668	0.1190	-0.01	-0.0356	-0.0219	-0.0004	-0.06	-0.0004	-0.0038	0.1114	0.9088	0.8819	
374	7	7	1.250	0.0036	0.0059	6.31	0.0002	0.0010	0.3237	0.1175	-0.01	-0.0330	-0.0798	-0.0014	-0.06	-0.0014	-0.0150	0.1093	0.9464	0.9288	
		5	0.0059	0.0002	0.0010	-0.0010	0.0010	0.1175	0.3237	0.1175	-0.01	-0.0330	-0.0798	-0.0014	-0.06	-0.0014	-0.0150	0.1093	0.9464	0.9288	

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																					
374	8	7	1.250	0.0033	0.0047	0.0001	0.0000	-0.00	0.2303	0.0455	-0.01	-2.71	0.1326	0.1347	0.0027	0.0023	-2.78	0.0247	0.0247	0.1051	0.0885	0.9315	0.9169
374	9	7	1.249	0.0020	0.0033	12.76	0.0000	-0.00	0.2175	0.1330	-0.02	-2.64	0.1770	0.1804	0.0036	0.0030	-2.73	0.0322	0.0322	0.1017	0.0851	0.9103	0.8933
374	10	7	1.250	0.0068	0.0037	-0.09	0.0003	-0.00	0.2678	0.1348	-0.00	-3.01	-0.0298	-0.0308	0.0010	-0.0009	-3.02	-0.0059	-0.0063	0.1752	0.1465	0.9952	1.0213
375	1	7	1.251	0.0048	0.0027	-3.17	0.0002	-0.00	0.2996	0.1092	-0.01	-0.16	-0.0537	-0.0519	0.0014	-0.0012	-0.10	-0.0102	-0.0099	0.1367	0.1171	0.9580	0.9616
375	2	7	1.251	0.0068	0.0037	-0.02	0.0003	-0.00	0.2726	0.0545	-0.00	-0.05	0.0040	0.0024	0.0000	0.0001	-0.02	0.0007	0.0004	0.1184	0.3275	0.9050	0.8167
375	3	7	1.249	0.0068	0.0040	1.05	0.0003	-0.00	0.2613	0.1278	-0.00	-0.02	0.0240	0.0204	0.0006	0.0006	0.00	0.0044	0.0038	0.1412	0.1643	0.9240	0.9498
375	4	7	1.250	0.0055	0.0045	3.18	0.0003	-0.00	0.2759	0.1454	-0.01	0.04	0.0614	0.0586	0.0013	0.0012	0.06	0.0115	0.0111	0.1097	0.1051	0.9414	0.9479
375	5	7	1.250	0.0035	0.0059	6.34	0.0002	0.00	0.2893	0.1315	-0.01	0.11	0.1154	0.1174	0.0023	0.0019	0.13	0.0218	0.0219	0.0996	0.0847	0.9449	0.9325
375	6	7	1.250	0.0031	0.0049	9.55	0.0001	-0.00	0.2402	0.0566	-0.01	0.20	0.1626	0.1652	0.0031	0.0027	0.19	0.0300	0.0300	0.0979	0.0638	0.9224	0.9082
375	7	7	1.249	0.0020	0.0032	12.80	0.0000	-0.00	0.1529	0.2300	-0.02	0.26	0.2002	0.2065	0.0038	0.0031	0.24	0.0317	0.0357	0.0972	0.0752	0.7936	0.8650
375	8	7	1.250	0.0068	0.0039	-0.03	0.0003	-0.00	0.2797	0.0984	-0.00	-0.07	0.0042	0.0025	0.0000	0.0001	-0.02	0.0006	0.0003	0.0329	0.2464	0.7419	0.5835

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key →																					
376	1	7	1.247	-5.15	-0.00	0.2700	-0.00	0.9714	0.92	-0.0012	0.05	0.88	0.1573	0.9527	0.0051	0.0002	0.0009	0.1550	0.8480	-0.0390	-0.0010	-0.0076	0.1297	0.9646
		6	1.249	-0.00	-0.0012	0.2796	-0.00	-0.9432	1.03	0.0004	0.05	0.98	0.1361	0.9307	0.0029	0.0003	0.0006	0.0690	0.8156	0.0135	0.0005	0.0025	0.2083	0.9331
		9	1.249	1.07	-0.00	0.2652	-0.00	-0.8924	1.07	0.0009	0.05	1.00	0.1312	0.9364	0.0038	0.0001	0.0006	0.1542	0.8499	0.0326	0.0009	0.0061	0.1448	0.9428
376	4	7	1.251	3.21	-0.00	0.2370	-0.00	-0.8604	1.13	0.0015	0.05	1.06	0.1026	0.9505	0.0053	0.0002	0.0009	0.2073	0.8531	0.0717	0.0013	0.0136	0.0971	0.9504
		6	1.250	6.39	-0.00	0.2572	-0.00	-0.8658	1.21	0.0024	0.06	1.13	0.0959	0.9414	0.0043	0.0001	0.0010	0.1713	0.8957	0.1285	0.0020	0.0238	0.0816	0.9292
		9	1.251	9.56	-0.00	0.2397	-0.00	-0.8221	1.29	0.0033	0.05	1.18	0.0953	0.9151	0.0043	0.0000	0.0008	0.1143	0.8210	0.1747	0.0028	0.0314	0.0819	0.9012
376	7	7	1.249	0.00	-0.0012	0.2778	-0.00	-0.9284	1.02	0.0004	0.06	0.98	0.1283	0.9131	0.0068	0.0003	0.0006	0.1414	0.9807	0.0126	0.0005	0.0023	0.2187	0.9291
		6	1.250	-3.10	-0.00	0.2634	-0.00	-0.9473	2.94	-0.0005	0.06	3.02	0.1782	0.9466	0.0026	0.0002	0.0004	0.1623	0.9071	-0.0133	-0.0003	-0.0027	0.1258	1.0355
		9	1.251	0.02	-0.0012	0.2729	-0.00	-0.9250	3.06	0.0010	0.05	3.11	0.1322	0.9516	0.0038	0.0000	0.0006	0.0535	0.7895	0.0395	0.0011	0.0075	0.1428	0.9498
377	3	7	1.249	1.11	-0.00	0.2456	-0.00	-0.8526	3.09	0.0014	0.06	3.13	0.1128	0.9600	0.0067	0.0003	0.0011	0.1311	0.8447	0.0595	0.0013	0.0114	0.1109	0.9583
		6	1.249	3.24	-0.00	0.2324	-0.00	-0.8301	3.15	0.0019	0.05	3.17	0.0949	0.9526	0.0037	0.0000	0.0006	0.2038	0.8864	0.0591	0.0016	0.0185	0.0838	0.9460
		9	1.251	0.0002	-0.0001	0.2038	-0.0007	0.8864	0.0981	0.0016	0.0016	0.0185	0.0838	0.9460	0.0040	0.0001	0.0007	0.2038	0.8864	0.0981	0.0016	0.0185	0.0838	0.9460

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key													
377	5	1.249 0.0039 0.0060	6.42 0.0001 0.0001	0.00 0.0006 0.0011	0.2539 0.1637	-0.01 0.8587 0.9496	3.22 0.1483 0.1507	0.06 0.0027 0.0024	3.24 0.277 0.0277	0.0937 0.0809	0.9348 0.9189				
377	6	1.249 0.0069 0.0032	0.02 0.0003 0.0001	-0.00 0.0012 0.0006	0.2599 0.1750	-0.00 0.9256 0.9313	3.06 0.0412 0.0392	0.05 0.0010 0.0011	3.12 0.077 0.0073	0.1271 0.1425	0.9428 0.9404				
378	1	1.249 0.0044 0.0030	-4.10 0.0002 0.0000	-0.00 0.0008 0.0005	0.3094 0.1148	-0.01 0.9903 0.8590	5.98 0.0007 0.0019	0.06 -0.0000 0.0002	6.00 0.0002 0.0004	-0.0707 0.7416	1.4018 1.0431				
378	2	1.249 0.0051 0.0031	-3.06 0.0002 0.0000	-0.00 0.0009 0.0004	0.2623 0.0842	-0.00 0.9457 0.7853	6.02 0.0187 0.0193	0.05 0.0006 0.0008	6.03 0.037 0.0037	0.1682 0.2079	0.9976 0.9825				
378	3	1.249 0.0069 0.0036	0.08 0.0003 0.0000	-0.00 0.0012 0.0006	0.2720 0.0339	-0.00 0.9096 0.8709	6.12 0.0786 0.0773	0.05 0.0016 0.0014	6.10 0.0153 0.0149	0.1032 0.0957	0.9743 0.9643				
378	4	1.250 0.0065 0.0037	1.15 0.0003 0.0000	-0.00 0.0011 0.0006	0.2683 0.0934	-0.00 0.8633 0.8627	6.15 0.0988 0.0966	0.05 0.0019 0.0016	6.13 0.0190 0.0185	0.0971 0.0858	0.9650 0.9585				
378	5	1.249 0.0053 0.0040	3.32 0.0002 0.0001	-0.00 0.0008 0.0007	0.2361 0.2201	-0.00 0.8207 0.9631	6.21 0.1329 0.1321	0.06 0.0024 0.0020	6.16 0.0251 0.0246	0.0928 0.0790	0.9453 0.9332				
378	6	1.248 0.0047 0.0046	4.38 0.0002 0.0002	-0.00 0.0007 0.0008	0.2206 0.2683	-0.01 0.8260 0.9499	6.23 0.1478 0.1488	0.05 0.0027 0.0023	6.19 0.0277 0.0274	0.0913 0.0785	0.9381 0.9229				
378	7	1.250 0.0072 0.0039	0.07 0.0003 0.0000	-0.00 0.0012 0.0007	0.2566 0.0622	-0.00 0.8903 0.8890	6.12 0.0790 0.0775	0.06 0.0016 0.0014	6.11 0.0153 0.0149	0.1020 0.0953	0.9698 0.9623				
379	1	1.248 0.0044 0.0031	-4.04 0.0002 0.0000	-0.00 0.0008 0.0005	0.2712 0.0791	-0.01 0.9274 0.9113	9.10 0.0360 0.0371	0.05 0.0010 0.0011	9.04 0.0072 0.0073	0.1483 0.1586	1.0091 0.9895				
379	2	1.248 0.0051 0.0033	-2.99 0.0002 0.0000	-0.00 0.0009 0.0005	0.2434 0.0261	-0.01 0.9405 0.8184	9.13 0.0359 0.0364	0.05 0.0013 0.0013	9.06 0.0111 0.0111	0.1191 0.1200	0.9949 0.9903				

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key										
379	3	7	1.248	0.16	-0.00	0.2613	-0.00	9.23	0.06	9.12	0.0923	0.9646
		8	0.0068	0.0003	0.0012	-0.0420	0.9044	0.1153	0.0021	0.0222	0.0805	0.9533
		9	0.0041	-0.0000	0.0007	0.0420	0.8630	0.1140	0.0018	0.0217		
379	4	7	1.248	1.19	-0.00	0.2677	-0.00	9.26	0.06	9.14	0.0949	0.9538
		8	0.0065	0.0003	0.0011	0.0757	0.8526	0.1324	0.0025	0.0252	0.0790	0.9405
		9	0.0040	0.0000	0.0007	0.0757	0.9169	0.1308	0.0020	0.0246		
379	10	7	1.248	2.23	-0.00	0.3344	-0.00	9.28	0.06	9.17	0.0981	0.9469
		8	0.0056	0.0003	0.0010	0.0427	0.9525	0.1473	0.0028	0.0279	0.0759	0.9298
		9	0.0036	0.0000	0.0007	0.0427	1.0873	0.1467	0.0022	0.0272		
379	11	7	1.248	0.11	-0.00	0.3353	-0.00	9.23	0.05	9.13	0.0985	0.9718
		8	0.0064	0.0004	0.0013	0.3353	1.0416	0.1131	0.0022	0.0219	0.0794	0.9556
		9	0.0038	-0.0001	0.0008	-0.1473	1.0529	0.1124	0.0017	0.0214		
380	1	7	1.250	-5.95	-0.00	0.3762	-0.01	15.12	0.06	15.09	0.1147	1.0275
		8	0.0034	0.0002	0.0007	0.0010	1.1382	0.0732	0.0016	0.0150	0.1046	0.9857
		9	0.0030	0.0000	0.0006	0.0010	1.0370	0.0760	0.0016	0.0152	0.0924	0.9857
380	2	7	1.251	-5.03	-0.00	0.4163	-0.01	15.15	0.06	15.11	0.0980	0.9929
		8	0.0036	0.0003	0.0008	0.0149	1.1657	0.0896	0.0018	0.0181	0.0793	0.9710
		9	0.0032	0.0000	0.0006	0.0149	1.0248	0.0932	0.0017	0.0183		
380	3	7	1.251	-3.91	-0.00	0.3773	-0.01	15.18	0.06	15.13	0.0938	0.9766
		8	0.0040	0.0003	0.0009	0.3773	1.1265	0.1082	0.0021	0.0214	0.0749	0.9581
		9	0.0033	-0.0000	0.0006	-0.1002	0.9563	0.1121	0.0017	0.0217		
380	4	7	1.250	-2.88	-0.00	0.3104	-0.00	15.21	0.06	15.15	0.0941	0.9620
		8	0.0047	0.0002	0.0010	0.3104	1.1054	0.1265	0.0023	0.0247	0.0714	0.9415
		9	0.0032	-0.0000	0.0006	-0.0864	0.9570	0.1300	0.0019	0.0249		
380	5	7	1.251	-1.87	-0.00	0.2866	-0.00	15.23	0.06	15.17	0.1918	0.9321
		8	0.0054	0.0003	0.0011	0.2866	1.0491	0.1441	0.0027	0.0277	0.1940	0.9025
		9	0.0029	-0.0000	0.0006	-0.0261	1.0842	0.1470	0.0021	0.0276		
381	1	7	0.901	-5.05	-0.00	0.4495	-0.01	15.16	0.06	15.15	0.2056	0.8795
		8	0.0042	0.0003	0.0009	0.4495	1.1248	0.1109	0.0042	0.0206	0.1980	0.8795
		9	0.0032	0.0002	0.0005	0.3344	0.7993	0.1132	0.0043	0.0204		
381	2	7	0.901	-3.91	-0.00	0.4232	-0.01	15.20	0.06	15.17	0.2056	0.8795
		8	0.0044	0.0003	0.0009	0.4232	1.0972	0.1258	0.0051	0.0227	0.1980	0.8795
		9	0.0031	0.0001	0.0004	0.2298	0.6936	0.1296	0.0051	0.0228		

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key													
381	5	0.901	-2.90	-0.00	0.3769	-0.01	15.22	0.06	15.19	0.2079	0.8670				
	6	0.0051	0.0003	0.0011	0.2091	0.0774	0.1405	0.0058	0.0249	0.2011	0.8668				
	9	0.0031	0.0001	0.0004	0.0004	0.6890	0.1436	0.0057	0.0247						
381	4	0.899	-1.93	-0.00	0.3425	-0.00	15.24	0.06	15.21	0.2036	0.8691				
	6	0.0066	0.0004	0.0014	0.2456	0.0612	0.1552	0.0063	0.0269	0.2030	0.8462				
	9	0.0037	0.0001	0.0005	0.0005	0.7673	0.1555	0.0063	0.0263						
381	5	0.899	-0.88	-0.00	0.3111	-0.00	15.27	0.06	15.21	0.1961	0.8621				
	6	0.0073	0.0004	0.0014	0.1904	0.9951	0.1717	0.0067	0.0296	0.1842	0.8400				
	9	0.0038	0.0001	0.0006	0.0006	0.8273	0.1717	0.0063	0.0288						
381	6	0.899	0.14	-0.00	0.3264	-0.00	15.27	0.06	15.22	0.1889	0.8488				
	6	0.0080	0.0005	0.0015	0.1087	0.9391	0.1815	0.0068	0.0308	0.1730	0.8351				
	9	0.0044	0.0000	0.0007	0.0007	0.7913	0.1836	0.0063	0.0306						
381	7	0.899	1.17	-0.00	0.3273	-0.00	15.29	0.06	15.23	0.1859	0.8375				
	6	0.0081	0.0005	0.0014	0.1264	0.9009	0.1916	0.0071	0.0321	0.1676	0.8280				
	9	0.0044	0.0001	0.0007	0.0007	0.8444	0.1933	0.0064	0.0320						
381	8	0.899	2.24	-0.00	0.3391	-0.00	15.30	0.06	15.24	0.1802	0.8270				
	6	0.0075	0.0005	0.0014	0.1611	0.9388	0.1994	0.0071	0.0329	0.1649	0.8130				
	9	0.0044	0.0001	0.0007	0.0007	0.8727	0.2011	0.0066	0.0327						
381	9	0.899	3.26	-0.00	0.3158	-0.00	15.30	0.06	15.24	0.1665	0.8230				
	6	0.0067	0.0004	0.0012	0.2163	0.9163	0.2082	0.0069	0.0342	0.1555	0.8109				
	9	0.0047	0.0002	0.0008	0.0008	0.8959	0.2071	0.0064	0.0335						
381	10	0.898	4.31	-0.00	0.2635	-0.01	15.29	0.06	15.23	0.1494	0.8217				
	6	0.0064	0.0003	0.0011	0.2803	0.8898	0.2132	0.0063	0.0350	0.1375	0.8085				
	9	0.0049	0.0002	0.0009	0.0009	0.9073	0.2129	0.0058	0.0344						
381	11	0.900	5.38	-0.00	0.2663	-0.01	15.29	0.07	15.22	0.1335	0.8205				
	6	0.0055	0.0002	0.0009	0.3055	0.8988	0.2160	0.0057	0.0354	0.1213	0.8076				
	9	0.0054	0.0003	0.0009	0.0009	0.8841	0.2148	0.0052	0.0347						
381	12	0.899	6.39	-0.00	0.2617	-0.01	15.29	0.07	15.22	0.1227	0.8192				
	6	0.0050	0.0002	0.0009	0.2769	0.9710	0.2198	0.0053	0.0360	0.1094	0.8058				
	9	0.0060	0.0003	0.0010	0.0010	0.8677	0.2201	0.0048	0.0354						
381	13	0.897	8.47	-0.00	0.1822	-0.01	15.29	0.06	15.22	0.1143	0.8180				
	6	0.0048	0.0001	0.0009	0.2524	0.9503	0.2295	0.0052	0.0375	0.0994	0.8035				
	9	0.0055	0.0002	0.0010	0.0010	0.9380	0.2300	0.0045	0.0369						

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key															
381	14	7	0.899	9.47	0.009	0.00	0.1961	-0.01	15.29	0.06	15.22	0.1122	0.8176					
		6	0.0048	0.0001	0.0009	0.0009	0.2100	0.9336	0.2312	0.0051	0.0378	0.0966	0.8017					
		5	0.0054	0.0002	0.0009	0.0009	0.2100	0.8778	0.2332	0.0045	0.0374							
381	15	7	0.894	10.55	-0.00	-0.00	0.2622	-0.01	15.29	0.06	15.22	0.1065	0.8162					
		6	0.0046	0.0002	0.0008	0.0008	0.1968	0.9361	0.2339	0.0049	0.0381	0.0881	0.8032					
		5	0.0048	0.0001	0.0008	0.0008	0.1968	0.8888	0.2362	0.0041	0.0379							
381	16	7	0.898	0.14	-0.00	-0.00	0.3207	-0.00	15.28	0.06	15.23	0.1866	0.8488					
		6	0.0082	0.0005	0.0015	0.0015	0.1645	0.9160	0.1833	0.0068	0.0311	0.1718	0.8353					
		5	0.0042	0.0001	0.0007	0.0007	0.1645	0.8991	0.1851	0.0063	0.0309							
382	1	7	0.901	-3.00	-0.00	-0.00	0.5020	-0.01	9.14	0.06	9.08	0.1851	0.9711					
		6	0.0042	0.0004	0.0010	0.0010	0.3727	1.1984	0.0719	0.0026	0.0139	0.1851	0.9581					
		5	0.0027	0.0002	0.0004	0.0004	0.3727	0.7423	0.0740	0.0027	0.0141							
382	2	7	0.901	0.09	-0.00	-0.00	0.3783	-0.00	9.25	0.06	9.15	0.2090	0.8926					
		6	0.0071	0.0005	0.0014	0.0014	0.2091	0.9871	0.1277	0.0053	0.0228	0.2020	0.8758					
		5	0.0039	0.0001	0.0006	0.0006	0.2091	0.8547	0.1267	0.0051	0.0222							
382	3	7	0.902	3.26	-0.00	-0.00	0.3584	-0.00	9.31	0.06	9.21	0.1964	0.8534					
		6	0.0065	0.0004	0.0012	0.0012	0.1975	0.9290	0.1722	0.0067	0.0294	0.1889	0.8391					
		5	0.0047	0.0001	0.0008	0.0008	0.1975	0.8963	0.1716	0.0064	0.0288							
382	4	7	0.900	6.37	-0.00	-0.00	0.2354	-0.01	9.35	0.07	9.23	0.1756	0.8279					
		6	0.0046	0.0002	0.0008	0.0008	0.2791	0.9269	0.2055	0.0072	0.0340	0.1624	0.8088					
		5	0.0063	0.0003	0.0011	0.0011	0.2791	0.8860	0.2065	0.0067	0.0334							
382	5	7	0.899	9.44	0.00	0.00	0.1897	-0.01	9.34	0.06	9.20	0.1262	0.8047					
		6	0.0045	0.0001	0.0008	0.0008	0.2288	0.9378	0.2255	0.0056	0.0370	0.1136	0.8210					
		5	0.0051	0.0002	0.0009	0.0009	0.2288	0.9648	0.2247	0.0051	0.0361							
382	7	7	0.899	10.55	-0.00	-0.00	0.2314	-0.01	9.33	0.07	9.20	0.1145	0.8192					
		6	0.0038	0.0001	0.0007	0.0007	0.1945	1.0257	0.2259	0.0051	0.0370	0.1043	0.8040					
		5	0.0047	0.0001	0.0008	0.0008	0.1945	0.9234	0.2262	0.0047	0.0363							
382	8	7	0.901	0.10	-0.00	-0.00	0.3893	-0.00	9.24	0.06	9.15	0.2102	0.8693					
		6	0.0072	0.0005	0.0014	0.0014	0.2509	1.0013	0.1280	0.0053	0.0227	0.2034	0.8733					
		5	0.0039	0.0001	0.0007	0.0007	0.2509	0.8883	0.1268	0.0051	0.0221							
383	1	7	0.901	-3.02	-0.00	-0.00	0.3937	-0.01	6.04	0.06	6.02	0.2458	1.0011					
		6	0.0049	0.0003	0.0010	0.0010	0.3259	1.0605	0.0267	0.0013	0.0053	0.2179	1.0145					
		5	0.0032	0.0002	0.0004	0.0004	0.3259	0.6763	0.0278	0.0012	0.0056							

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key															
383	2	7	0.899	0.005	-0.00	0.4280	-0.00	6.14	0.06	6.11	0.1995	0.9403						
		8	0.0068	0.0005	0.0013	0.2135	1.0011	0.0942	0.0037	0.0177	0.1986	0.9204						
		9	0.0041	0.0001	0.0006		0.7425	0.0914	0.0036	0.0168								
383	3	7	0.900	1.10	-0.00	0.4127	-0.00	6.17	0.06	6.13	0.2135	0.9050						
		8	0.0072	0.0006	0.0013	0.1290	0.9560	0.1116	0.0047	0.0202	0.2116	0.8928						
		9	0.0044	0.0001	0.0006		0.7823	0.1082	0.0045	0.0193								
383	4	7	0.900	3.21	-0.00	0.3995	-0.00	6.22	0.06	6.18	0.2114	0.8684						
		8	0.0061	0.0004	0.0011	0.1722	0.9240	0.1441	0.0060	0.0250	0.2098	0.8532						
		9	0.0048	0.0001	0.0008		0.8763	0.1421	0.0059	0.0242								
383	5	7	0.898	6.34	0.00	0.3086	-0.01	6.27	0.07	6.23	0.1898	0.8418						
		8	0.0043	0.0002	0.0008	0.2626	1.0100	0.1845	0.0070	0.0310	0.1792	0.8231						
		9	0.0065	0.0003	0.0010		0.8381	0.1867	0.0066	0.0307								
383	6	7	0.899	9.44	0.00	0.1729	-0.01	6.30	0.06	6.24	0.1595	0.8042						
		8	0.0042	0.0001	0.0008	0.1907	1.0278	0.2164	0.0069	0.0356	0.1454	0.8226						
		9	0.0060	0.0002	0.0010		0.8620	0.2175	0.0063	0.0349								
383	7	7	0.900	12.62	-0.00	0.2118	-0.01	6.28	0.06	6.21	0.1109	0.8026						
		8	0.0034	0.0001	0.0006	0.1838	0.9487	0.2298	0.0051	0.0376	0.1034	0.8197						
		9	0.0035	0.0001	0.0006		0.9147	0.2303	0.0047	0.0369								
383	8	7	0.901	0.07	-0.00	0.4373	-0.00	6.13	0.06	6.11	0.2025	0.9195						
		8	0.0068	0.0005	0.0013	0.2144	0.9961	0.0929	0.0037	0.0172	0.1979	0.9290						
		9	0.0039	0.0001	0.0006		0.8405	0.0912	0.0036	0.0167								
384	1	7	0.898	-3.07	-0.00	0.3918	-0.01	2.55	0.06	3.01	0.3138	0.9561						
		8	0.0048	0.0003	0.0010	0.3698	1.0768	-0.0119	-0.0007	-0.0022	0.3603	0.9249						
		9	0.0025	0.0001	0.0004		0.8427	-0.0101	-0.0007	-0.0018								
384	2	7	0.898	0.00	-0.00	0.4806	-0.00	3.06	0.06	3.10	0.2469	0.9453						
		8	0.0065	0.0006	0.0012	0.2547	0.9771	0.0479	0.0023	0.0090	0.2253	0.9353						
		9	0.0038	0.0001	0.0005		0.7077	0.0474	0.0021	0.0088								
384	3	7	0.900	1.07	-0.00	0.4642	-0.00	3.09	0.07	3.13	0.2286	0.9378						
		8	0.0070	0.0006	0.0013	0.1901	0.9575	0.0713	0.0032	0.0133	0.2103	0.9331						
		9	0.0044	0.0001	0.0007		0.8179	0.0698	0.0029	0.0130								
384	4	7	0.901	3.18	-0.00	0.4392	-0.00	3.16	0.06	3.19	0.2130	0.8964						
		8	0.0061	0.0005	0.0010	0.1715	0.8890	0.1122	0.0047	0.0201	0.2133	0.8867						
		9	0.0049	0.0001	0.0008		0.8622	0.1093	0.0046	0.0193								

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

← See Key →

Run	Pt.	Cd																				
384	5	7	0.877	6.30	0.00	0.2904	-0.01	3.23	0.07	3.27	0.2033	0.6509										
	6	8	0.0046	0.0002	0.0008	0.2535	0.9336	0.1593	0.0064	0.0273	0.2009	0.8378										
	9	9	0.0069	0.0003	0.0011	0.0000	0.8176	0.1608	0.0064	0.0269												
384	6	7	0.898	9.41	0.00	0.2403	-0.01	3.28	0.06	3.30	0.1858	0.8295										
	8	8	0.0039	0.0001	0.0008	0.2213	1.0436	0.1977	0.0073	0.0328	0.1717	0.8113										
	9	9	0.0058	0.0002	0.0011	0.0000	0.9390	0.1995	0.0068	0.0323												
384	7	7	0.897	12.61	0.00	0.2608	-0.01	3.28	0.06	3.29	0.1335	0.8208										
	8	8	0.0033	0.0001	0.0006	0.1753	0.9717	0.2259	0.0060	0.0370	0.1219	0.8040										
	9	9	0.0038	0.0001	0.0006	0.0000	0.8801	0.2267	0.0055	0.0364												
384	8	7	0.897	0.01	-0.00	0.4929	-0.00	3.05	0.06	3.11	0.2445	0.9356										
	8	8	0.0065	0.0006	0.0013	0.3306	1.0100	0.0483	0.0023	0.0090	0.2267	0.9334										
	9	9	0.0038	0.0002	0.0006	0.0000	0.8291	0.0480	0.0021	0.0089												
385	1	7	0.898	-5.10	-0.00	0.3907	-0.01	0.93	0.06	0.89	0.2131	0.9378										
	8	8	0.0049	0.0003	0.0010	0.3654	1.0448	-0.0425	-0.0018	-0.0079	0.1881	0.9450										
	9	9	0.0025	0.0001	0.0004	0.0000	0.8783	-0.0419	-0.0015	-0.0079												
385	2	7	0.897	-0.02	-0.00	0.4710	-0.00	1.05	0.06	0.97	0.3478	0.9529										
	8	8	0.0063	0.0006	0.0012	0.3141	1.0034	0.1187	0.0013	0.0035	0.2785	0.9419										
	9	9	0.0037	0.0002	0.0005	0.0000	0.7254	0.0167	0.0009	0.0031												
385	3	7	0.899	1.05	-0.00	0.5519	-0.00	1.09	0.06	1.00	0.2601	0.9314										
	8	8	0.0064	0.0007	0.0013	0.2086	1.0082	0.0417	0.0021	0.0077	0.2285	0.9396										
	9	9	0.0041	0.0001	0.0006	0.0000	0.7564	0.0375	0.0017	0.0070												
385	4	7	0.897	3.17	-0.00	0.4568	-0.00	1.16	0.06	1.06	0.2203	0.9214										
	8	8	0.0057	0.0005	0.0010	0.1318	0.9043	0.0870	0.0038	0.0160	0.2086	0.9190										
	9	9	0.0052	0.0001	0.0008	0.0000	0.7852	0.0837	0.0034	0.0153												
385	5	7	0.898	6.31	0.00	0.3700	-0.01	1.23	0.07	1.14	0.2118	0.8710										
	8	8	0.0040	0.0002	0.0007	0.2377	0.9801	0.1393	0.0059	0.0238	0.2133	0.8543										
	9	9	0.0067	0.0003	0.0011	0.0000	0.8319	0.1392	0.0059	0.0238												
385	6	7	0.899	9.44	0.00	0.2752	-0.01	1.29	0.06	1.19	0.1907	0.8414										
	8	8	0.0037	0.0002	0.0007	0.1869	0.9945	0.1837	0.0070	0.0309	0.1819	0.8236										
	9	9	0.0061	0.0002	0.0010	0.0000	0.8655	0.1851	0.0067	0.0305												
385	7	7	0.901	12.58	0.00	0.1693	-0.01	1.32	0.06	1.21	0.1571	0.8051										
	8	8	0.0034	0.0001	0.0006	0.1553	0.9257	0.2205	0.0069	0.0363	0.1442	0.8054										
	9	9	0.0035	0.0001	0.0006	0.0000	0.8816	0.2214	0.0063	0.0356												

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																		
385	8	0.899	-0.002	-0.000	0.4836	-0.00	1.04	0.06	0.97	0.3134	0.9299	0.0066	0.0006	0.0013	0.0127	0.0193	0.0012	0.0035	0.2639	0.9171
		0.0036	0.0002	0.0006	0.4044	0.8292	0.0170	0.0009	0.0031											
386	1	0.901	-5.10	-0.00	0.4566	-0.01	-0.16	0.06	-0.11	0.2090	0.9354	0.0044	0.0004	0.0010	1.1634	-0.0588	-0.0024	-0.0110	0.1953	0.9486
		0.0028	0.0001	0.0004	0.2681	0.8084	-0.0564	-0.0022	-0.0107											
386	2	0.899	-0.04	-0.00	0.4638	-0.00	-0.06	0.06	-0.03	0.4550	0.8602	0.0066	0.0005	0.0013	1.0060	0.0051	0.0004	0.0008	0.4143	0.6279
		0.0039	0.0002	0.0006	0.2854	0.6400	0.0034	0.0002	0.0004											
386	3	0.898	3.13	-0.00	0.5031	-0.00	0.05	0.06	0.06	0.2349	0.9211	0.0057	0.0010	0.0010	0.9264	0.0705	0.0033	0.0129	0.2134	0.9055
		0.0049	0.0001	0.0008	0.1496	0.8512	0.0684	0.0029	0.0123											
386	4	0.899	6.26	0.00	0.3629	-0.01	0.13	0.07	0.14	0.2055	0.8810	0.0039	0.0007	0.0010	0.9366	0.1297	0.0053	0.0228	0.2053	0.8556
		0.0064	0.0003	0.0010	0.2562	0.8415	0.1309	0.0053	0.0224											
386	5	0.899	9.41	0.00	0.2477	-0.01	0.20	0.07	0.19	0.1892	0.8508	0.0041	0.0007	0.0011	0.9612	0.1769	0.0066	0.0301	0.1854	0.8258
		0.0061	0.0002	0.0011	0.2256	0.9254	0.1770	0.0065	0.0292											
386	6	0.898	12.59	0.00	0.1705	-0.02	0.24	0.06	0.23	0.1682	0.8265	0.0036	0.0006	0.0006	0.8846	0.2138	0.0071	0.0353	0.1537	0.8066
		0.0040	0.0000	0.0006	0.1175	0.7479	0.2146	0.0066	0.0346											
386	7	0.897	-0.04	-0.00	0.4424	-0.00	-0.05	0.06	-0.02	0.4120	0.7960	0.0067	0.0005	0.0005	0.9691	0.0051	0.0004	0.0008	0.4130	0.5775
		0.0035	0.0002	0.0005	0.3859	0.7442	0.0033	0.0002	0.0003											
387	1	0.899	-5.16	-0.00	0.4080	-0.01	-3.10	0.06	-3.10	0.2078	0.9085	0.0045	0.0009	0.0004	1.0678	-0.1014	-0.0042	-0.0184	0.2025	0.9309
		0.0028	0.0000	0.0004	0.1455	0.7455	-0.0995	-0.0040	-0.0185											
387	2	0.899	-0.08	-0.00	0.4354	-0.00	-2.99	0.06	-3.00	0.1690	0.9672	0.0066	0.0013	0.0005	1.0021	-0.0349	-0.0011	-0.0067	0.1893	0.9925
		0.0034	0.0002	0.0005	0.3603	0.7898	-0.0360	-0.0013	-0.0071											
387	3	0.899	3.09	-0.00	0.5373	-0.00	-2.89	0.06	-2.92	0.3081	0.8664	0.0054	0.0010	0.0006	0.9264	0.0257	0.0015	0.0045	0.2295	0.8664
		0.0048	0.0001	0.0006	0.1340	0.7210	0.0234	0.0010	0.0040											

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																					
387	4	7	0.899	0.0036	0.0067	6.25	0.0003	0.0002	0.00	0.0007	0.4240	0.2198	-0.01	-0.9846	0.8229	-2.70	0.0919	0.0039	0.0038	-2.82	0.0167	0.0168	0.2140	0.8914
387	5	7	0.900	0.0035	0.0065	9.36	0.0002	0.0002	0.00	0.0007	0.3155	0.1841	-0.01	1.0106	0.8285	-2.70	0.1494	0.0060	0.0059	-2.74	0.0259	0.0252	0.2008	0.8693
387	6	7	0.899	0.0031	0.0036	12.58	0.0001	0.0001	0.00	0.0005	0.2983	0.1824	-0.01	0.9134	0.8941	-2.64	0.1952	0.0068	0.0064	-2.70	0.0329	0.0322	0.1750	0.8433
387	7	7	0.898	0.0069	0.0032	-0.07	0.0006	0.0002	-0.00	0.0013	0.4382	0.4431	-0.00	1.0020	0.8632	-2.99	-0.0346	0.0012	0.0013	-3.00	0.0068	-0.0072	0.1781	0.9892
388	3	7	0.900	-0.0827	-0.0826	-3.24	-0.0043	-0.0032	45.00	-0.0152	0.2646	0.1970	-3.13	0.9226	0.9425	-3.09	-0.0825	-0.0043	-0.0029	-3.07	-0.0152	-0.0154	0.2604	0.9227
388	4	7	0.900	-0.0377	-0.0377	-0.09	-0.0017	-0.0014	45.00	-0.0070	0.2351	0.1928	-3.04	0.9377	0.9486	-3.00	-0.0398	-0.0016	-0.0011	-3.00	-0.0077	-0.0074	0.2026	0.9667
388	5	7	0.899	-0.0196	-0.0214	0.96	-0.0013	-0.0009	44.99	-0.0035	0.3378	0.2096	-3.02	0.9177	0.9705	-2.97	-0.0257	-0.0009	-0.0008	-2.99	-0.0052	-0.0045	0.1763	1.0162
388	6	7	0.899	0.0039	0.0078	5.07	0.0007	0.0005	44.99	0.0008	0.8943	0.3483	-2.96	1.0986	0.8690	-2.93	0.0008	0.0001	0.0001	-2.95	-0.0000	-0.0005	0.6213	-0.1650
388	7	7	0.898	0.0492	0.0581	6.19	0.0024	0.0029	44.99	0.0089	0.2470	0.2497	-2.90	0.9049	0.9423	-2.86	0.0435	0.0020	0.0023	-2.89	0.0075	0.0065	0.2319	0.8722
388	8	7	0.900	0.0933	0.1010	9.35	0.0037	0.0055	44.99	0.0171	0.2028	0.2760	-2.83	0.9176	0.9276	-2.79	0.0880	0.0054	0.0054	-2.80	0.0159	0.0163	0.1951	0.9061
388	9	7	0.901	0.1347	0.1323	12.54	0.0045	0.0073	44.99	0.0243	0.1689	0.2779	-2.78	0.9031	0.8663	-2.75	0.1322	0.0039	0.0072	-2.75	0.0240	0.0219	0.1494	0.9102

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key															
388	10	7	0.898	-0.0017	45.00	0.2335	-3.04	-3.00	-3.03	-3.00	-3.00	-3.00	-3.00	-3.00	-3.00	-3.00	0.2114	0.9943
		6	-0.0375	-0.0013	-0.0071	0.1759	0.9491	-0.0389	-0.0016	-0.0011	-0.0077	-0.0075	-0.0075	-0.0075	-0.0075	-0.0075	0.1508	1.0153
		5	-0.0381	-0.0013	-0.0071	0.1759	0.9325	-0.0371	-0.0011	-0.0011	-0.0075	-0.0075	-0.0075	-0.0075	-0.0075	-0.0075		
389	1	7	0.898	-0.0015	44.99	0.2420	-0.09	-0.13	-0.00	-0.00	-0.07	-0.07	-0.07	-0.07	-0.07	0.2557	0.9503	
		6	-0.0407	-0.0019	-0.0075	0.1897	0.9258	-0.0403	-0.0020	-0.0012	-0.0076	-0.0073	-0.0073	-0.0073	-0.0073	-0.0073	0.1652	0.9564
		5	-0.0396	-0.0015	-0.0071	0.1897	0.9057	-0.0385	-0.0012	-0.0012	-0.0073	-0.0073	-0.0073	-0.0073	-0.0073	-0.0073		
389	2	7	0.899	-0.001	44.99	0.7065	-0.01	-0.06	0.07	0.07	-0.03	-0.03	-0.03	-0.03	-0.03	0.0491	0.5148	
		6	0.0023	0.0003	0.0007	0.6765	1.5064	0.0016	0.0000	0.0002	0.0001	0.0001	0.0001	0.0001	0.0001	0.4354	0.5827	
		5	0.0037	0.0005	0.0008	0.6765	1.1428	0.0033	0.0002	0.0002	0.0003	0.0003	0.0003	0.0003	0.0003			
389	3	7	0.898	0.0012	44.99	0.3765	0.00	-0.03	0.10	0.10	-0.00	-0.00	-0.00	-0.00	-0.00	0.3520	0.9135	
		6	0.0167	0.0012	0.0033	0.3898	0.9895	0.0127	0.0008	0.0010	0.0023	0.0023	0.0023	0.0023	0.0023	0.3147	0.8898	
		5	0.0174	0.0013	0.0034	0.3898	0.9797	0.0162	0.0010	0.0010	0.0023	0.0023	0.0023	0.0023	0.0023			
389	4	7	0.899	0.0021	44.99	0.2054	0.05	-0.00	0.16	0.16	0.04	0.04	0.04	0.04	0.04	0.1982	0.9105	
		6	0.0513	0.0021	0.0096	0.2449	0.9408	0.0412	0.0016	0.0022	0.0075	0.0075	0.0075	0.0075	0.0075	0.2531	0.9213	
		5	0.0536	0.0026	0.0101	0.2449	0.9456	0.0453	0.0022	0.0022	0.0063	0.0063	0.0063	0.0063	0.0063			
389	5	7	0.897	0.0038	44.99	0.2034	0.12	0.07	0.25	0.25	0.12	0.12	0.12	0.12	0.12	0.1927	0.9242	
		6	0.0943	0.0038	0.0175	0.2819	0.9300	0.0861	0.0033	0.0049	0.159	0.159	0.159	0.159	0.159	0.2616	0.8847	
		5	0.0986	0.0055	0.0179	0.2819	0.9123	0.0875	0.0049	0.0049	0.154	0.154	0.154	0.154	0.154			
389	6	7	0.897	0.0042	44.99	0.1696	0.17	0.11	0.31	0.31	0.18	0.18	0.18	0.18	0.18	0.1545	0.9138	
		6	0.1352	0.0045	0.0245	0.2789	0.9064	0.1299	0.0040	0.0072	0.237	0.237	0.237	0.237	0.237	0.2882	0.8418	
		5	0.1312	0.0073	0.0229	0.2789	0.8735	0.1256	0.0072	0.0072	0.211	0.211	0.211	0.211	0.211			
389	7	7	0.899	0.0058	44.99	0.1442	0.21	0.16	0.36	0.36	0.21	0.21	0.21	0.21	0.21	0.1331	0.9040	
		6	0.1735	0.0050	0.0310	0.2617	0.8955	0.1689	0.0044	0.0078	0.305	0.305	0.305	0.305	0.305	0.2468	0.8123	
		5	0.1590	0.0083	0.0266	0.2617	0.8365	0.1582	0.0078	0.0078	0.257	0.257	0.257	0.257	0.257			
389	8	7	0.896	-0.000	44.99	0.5131	-0.01	-0.06	0.05	0.05	-0.02	-0.02	-0.02	-0.02	-0.02	0.0412	0.3674	
		6	0.0028	0.0002	0.0007	0.6371	1.3434	0.0019	0.0000	0.0003	0.0001	0.0001	0.0001	0.0001	0.0001	0.4142	0.4675	
		5	0.0039	0.0005	0.0008	0.6371	1.0131	0.0037	0.0003	0.0003	0.0003	0.0003	0.0003	0.0003	0.0003			
390	1	7	0.898	-0.0013	44.99	0.2774	0.96	0.96	0.96	0.96	0.92	0.92	0.92	0.92	0.92	0.2521	0.9809	
		6	-0.0247	-0.0013	-0.0044	0.2095	0.8925	-0.0241	-0.0012	-0.0012	-0.0047	-0.0047	-0.0047	-0.0047	-0.0047	0.1563	0.9881	
		5	-0.0243	-0.0010	-0.0043	0.2095	0.9025	-0.0241	-0.0007	-0.0007	-0.0047	-0.0047	-0.0047	-0.0047	-0.0047			
390	2	7	0.897	-0.000	44.99	0.3921	1.04	1.04	1.04	1.04	0.98	0.98	0.98	0.98	0.98	0.3577	0.9373	
		6	0.0149	0.0011	0.0031	0.3940	1.0416	0.0134	0.0009	0.0009	0.0025	0.0025	0.0025	0.0025	0.0025	0.3104	0.9159	
		5	0.0159	0.0012	0.0031	0.3940	0.9794	0.0157	0.0009	0.0009	0.0028	0.0028	0.0028	0.0028	0.0028			

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Pt.	Cd	See Key															
390	5	7	0.897	1.05	44.99	0.0013	0.0017	1.05	0.9850	1.05	0.262	0.0308	1.07	0.0012	1.01	0.2302	0.9351	
		6	0.0347	0.0013	0.0068	0.0017	0.0017	0.1967	0.9628	0.0282	0.0308	0.0016	0.0052	0.0057	0.2592	0.9275		
		5	0.0340	0.0017	0.0065	0.0017	0.0017	0.2524	0.9628	0.0308	0.0308	0.0016	0.0057	0.0057	0.2592	0.9275		
390	4	7	0.898	3.16	44.99	0.0026	0.0035	1.11	0.9437	1.09	0.0574	1.13	1.05	0.1838	0.9271			
		6	0.0674	0.0026	0.0127	0.0026	0.0035	0.1951	0.9437	0.0574	0.0597	0.0031	0.0106	0.0109	0.2660	0.9195		
		5	0.0675	0.0035	0.0127	0.0035	0.0035	0.2627	0.9437	0.0597	0.0597	0.0031	0.0106	0.0109	0.2660	0.9195		
390	5	7	0.897	6.29	44.99	0.0041	0.0062	1.17	0.8960	1.16	0.1026	1.22	1.13	0.1764	0.9274			
		6	0.1081	0.0041	0.0199	0.0041	0.0062	0.1932	0.8960	0.1026	0.0992	0.0036	0.0119	0.0174	0.2884	0.8787		
		5	0.1093	0.0062	0.0196	0.0062	0.0196	0.2848	0.8960	0.0992	0.0992	0.0036	0.0119	0.0174	0.2884	0.8787		
390	6	7	0.898	9.44	44.99	0.0047	0.0077	1.22	0.9027	1.21	0.1437	1.28	1.19	0.1486	0.9072			
		6	0.1475	0.0047	0.0266	0.0047	0.0077	0.1608	0.9027	0.1437	0.1355	0.0042	0.0226	0.0266	0.2838	0.8360		
		5	0.1401	0.0077	0.0241	0.0077	0.0241	0.2778	0.8624	0.1355	0.1355	0.0076	0.0226	0.0226	0.2838	0.8360		
390	7	7	0.898	12.60	44.99	0.0051	0.0084	1.26	0.8911	1.25	0.1830	1.32	1.21	0.1268	0.8989			
		6	0.1851	0.0051	0.0329	0.0051	0.0084	0.1385	0.8911	0.1830	0.1660	0.0046	0.0229	0.0268	0.2413	0.8072		
		5	0.1657	0.0084	0.0273	0.0084	0.0273	0.2550	0.8256	0.1660	0.1660	0.0080	0.0268	0.0268	0.2413	0.8072		
390	8	7	0.896	-0.00	44.99	0.0011	0.0031	1.04	0.9887	1.03	0.0135	1.04	0.98	0.3506	0.9084			
		6	0.0153	0.0011	0.0030	0.0011	0.0031	0.3750	0.9887	0.0135	0.0157	0.0009	0.0024	0.0028	0.3199	0.8982		
		5	0.0162	0.0031	0.0031	0.0031	0.0031	0.4045	0.9661	0.0157	0.0157	0.0010	0.0028	0.0028	0.3199	0.8982		
391	1	7	0.895	-3.08	44.99	0.0003	0.0004	3.02	0.9608	2.98	0.0029	2.92	3.05	0.2039	1.0161			
		6	0.0005	0.0003	0.0004	0.0003	0.0004	2.6603	4.1808	0.0029	0.0035	0.0001	0.0004	0.0006	0.3734	0.6274		
		5	0.0008	0.0001	0.0004	0.0001	0.0004	1.1599	2.7362	0.0035	0.0035	0.0002	0.0004	0.0004	0.3734	0.6274		
391	2	7	0.896	0.05	44.99	0.0018	0.0022	3.09	0.9833	3.05	0.0440	3.00	3.12	0.2043	0.9399			
		6	0.0486	0.0018	0.0095	0.0018	0.0022	0.1924	0.9833	0.0440	0.0465	0.0017	0.0082	0.0085	0.2572	0.9142		
		5	0.0459	0.0022	0.0088	0.0022	0.0088	0.2441	0.9642	0.0465	0.0465	0.0023	0.0085	0.0085	0.2572	0.9142		
391	3	7	0.898	1.09	44.99	0.0024	0.0030	3.12	0.9647	3.07	0.0586	3.04	3.15	0.1818	0.9382			
		6	0.0664	0.0024	0.0128	0.0024	0.0030	0.1831	0.9647	0.0586	0.0616	0.0021	0.0110	0.0112	0.2585	0.9124		
		5	0.0631	0.0030	0.0120	0.0030	0.0120	0.2432	0.9548	0.0616	0.0616	0.0031	0.0112	0.0112	0.2585	0.9124		
391	4	7	0.898	3.20	44.99	0.0038	0.0049	3.17	0.9433	3.11	0.0865	3.10	3.20	0.1758	0.9455			
		6	0.0955	0.0038	0.0173	0.0038	0.0049	0.1996	0.9433	0.0865	0.0893	0.0030	0.0159	0.0159	0.2583	0.8915		
		5	0.0941	0.0049	0.0173	0.0049	0.0173	0.2631	0.9198	0.0893	0.0893	0.0046	0.0159	0.0159	0.2583	0.8915		
391	5	7	0.898	6.34	44.99	0.0046	0.0071	3.23	0.9058	3.16	0.1296	3.17	3.26	0.1580	0.9217			
		6	0.1352	0.0046	0.0245	0.0046	0.0071	0.1735	0.9058	0.1296	0.1234	0.0040	0.0238	0.0210	0.2789	0.8535		
		5	0.1282	0.0071	0.0226	0.0071	0.0226	0.2800	0.8817	0.1234	0.1234	0.0068	0.0210	0.0210	0.2789	0.8535		

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																					
391	6	7	0.897	9.47	44.99	0.1447	0.8974	3.27	3.21	3.22	3.30	0.1355	0.9068	0.1734	0.0050	0.0311	0.02627	0.8498	0.1697	0.0046	0.0307	0.2551	0.8200	
		6	0.1582	0.0083	0.0268																			
391	7	7	0.897	12.63	44.99	0.1324	0.8841	3.30	3.25	3.25	3.31	0.1161	0.8994	0.2067	0.0054	0.0365	0.2326	0.8147	0.2066	0.0047	0.0371	0.2200	0.7906	
		6	0.1780	0.0082	0.0290																			
391	8	7	0.898	0.03	44.99	0.1888	0.9785	3.09	3.05	3.01	3.12	0.1923	0.9247	0.0487	0.0018	0.0095	0.2399	0.9615	0.0468	0.0017	0.0085	0.2466	0.9105	
		6	0.0463	0.0022	0.0089																			
392	1	7	0.897	-3.01	44.99	0.2109	1.0049	6.09	6.07	6.10	6.06	0.2167	0.9602	0.0465	0.0019	0.0089	0.1857	0.9837	0.0455	0.0017	0.0083	0.1950	0.9184	
		6	0.0448	0.0016	0.0088																			
392	2	7	0.896	0.11	44.99	0.1926	0.9552	6.15	6.13	6.19	6.13	0.1727	0.9573	0.0938	0.0036	0.0179	0.2318	0.9370	0.0892	0.0041	0.0159	0.2303	0.8929	
		6	0.0905	0.0042	0.0169																			
392	3	7	0.896	1.15	44.99	0.1936	0.9114	6.18	6.15	6.22	6.16	0.1670	0.9475	0.1086	0.0042	0.0203	0.2469	0.9351	0.1026	0.0034	0.0194	0.2483	0.8790	
		6	0.1035	0.0051	0.0188																			
392	4	7	0.898	3.25	44.99	0.1790	0.9109	6.21	6.18	6.27	6.19	0.1552	0.9254	0.1355	0.0048	0.0223	0.2645	0.8815	0.1228	0.0039	0.0210	0.2561	0.8566	
		6	0.1266	0.0067	0.0223																			
392	5	7	0.896	6.38	44.99	0.1562	0.8956	6.26	6.23	6.33	6.24	0.1382	0.9089	0.1721	0.0053	0.0308	0.2592	0.8532	0.1696	0.0046	0.0308	0.2544	0.8297	
		6	0.1593	0.0082	0.0271																			
392	6	7	0.897	9.52	44.99	0.1404	0.8780	6.29	6.27	6.36	6.27	0.1250	0.8983	0.2034	0.0057	0.0357	0.2343	0.8232	0.2055	0.0049	0.0369	0.2250	0.7985	
		6	0.1798	0.0084	0.0296																			
392	7	7	0.896	12.66	44.99	0.1274	0.8625	6.32	6.30	6.33	6.25	0.1188	0.8769	0.2308	0.0058	0.0398	0.1733	0.8125	0.1979	0.0055	0.0313	0.1635	0.7908	
		6	0.1933	0.0067	0.0314																			
392	8	7	0.895	0.11	44.99	0.1894	0.9573	6.15	6.14	6.19	6.13	0.1697	0.9532	0.0947	0.0035	0.0181	0.2243	0.9330	0.0898	0.0039	0.0171	0.2199	0.9067	
		6	0.0916	0.0041	0.0171																			

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd																		
393	1	7	0.897	-2.94	44.99	0.1970	0.9655	9.18	9.12	9.10	0.1708	0.9653								
		8	0.0924	0.0036	0.0178	0.1875	0.9452	0.0939	0.0032	0.0181	0.1867	0.9219								
		9	0.0876	0.0032	0.0165	0.1875	0.9452	0.0886	0.0033	0.0163	0.1867	0.9219								
393	2	7	0.895	0.18	44.99	0.1846	0.9190	9.23	9.21	9.17	0.1649	0.9233								
		8	0.1333	0.0049	0.0245	0.2364	0.8903	0.1306	0.0043	0.0241	0.2336	0.8590								
		9	0.1238	0.0058	0.0220	0.2364	0.8903	0.1243	0.0058	0.0213	0.2336	0.8590								
393	3	7	0.896	1.20	44.99	0.1802	0.923	9.24	9.23	9.16	0.1563	0.9186								
		8	0.1459	0.0052	0.0265	0.2470	0.8809	0.1429	0.0044	0.0262	0.2384	0.8517								
		9	0.1347	0.0066	0.0237	0.2470	0.8809	0.1339	0.0063	0.0228	0.2384	0.8517								
393	4	7	0.896	3.30	44.99	0.1707	0.8666	9.27	9.28	9.21	0.1436	0.9097								
		8	0.1691	0.0057	0.0299	0.2590	0.8549	0.1677	0.0048	0.0305	0.2460	0.8296								
		9	0.1544	0.0080	0.0264	0.2590	0.8549	0.1525	0.0075	0.0253	0.2460	0.8296								
393	5	7	0.896	6.42	44.99	0.1501	0.8717	9.31	9.31	9.24	0.1287	0.8954								
		8	0.2005	0.0060	0.0349	0.2300	0.8317	0.2049	0.0052	0.0367	0.2168	0.8151								
		9	0.1815	0.0083	0.0302	0.2300	0.8317	0.1818	0.0078	0.0296	0.2168	0.8151								
393	6	7	0.895	9.55	44.99	0.1284	0.8638	9.34	9.32	9.23	0.1210	0.8780								
		8	0.2305	0.0059	0.0398	0.1878	0.8170	0.2322	0.0056	0.0407	0.1794	0.7962								
		9	0.1977	0.0074	0.0323	0.1878	0.8170	0.1995	0.0071	0.0317	0.1794	0.7962								
393	7	7	0.895	0.17	44.99	0.1848	0.9170	9.22	9.21	9.17	0.1630	0.9198								
		8	0.1333	0.0049	0.0244	0.2366	0.8896	0.1310	0.0042	0.0241	0.2353	0.8618								
		9	0.1236	0.0058	0.0220	0.2366	0.8896	0.1236	0.0058	0.0213	0.2353	0.8618								
394	1	7	0.896	-2.84	44.99	0.1913	0.8928	15.25	15.25	15.20	0.1705	0.8982								
		8	0.1646	0.0063	0.0294	0.1994	0.8781	0.1677	0.0057	0.0301	0.1872	0.8428								
		9	0.1530	0.0061	0.0268	0.1994	0.8781	0.1568	0.0058	0.0264	0.1872	0.8428								
394	2	7	0.899	0.27	44.99	0.1596	0.8560	15.27	15.30	15.24	0.1402	0.8937								
		8	0.1997	0.0063	0.0351	0.2038	0.8794	0.2030	0.0056	0.0362	0.1994	0.8242								
		9	0.1831	0.0074	0.0313	0.2038	0.8560	0.1828	0.0072	0.0301	0.1994	0.8242								
394	3	7	0.895	1.29	44.99	0.1533	0.8730	15.29	15.31	15.25	0.1381	0.8839								
		8	0.2093	0.0064	0.0365	0.2022	0.8475	0.2111	0.0058	0.0373	0.1963	0.8158								
		9	0.1915	0.0077	0.0324	0.2022	0.8475	0.1893	0.0074	0.0308	0.1963	0.8158								
394	4	7	0.897	2.32	44.99	0.1437	0.8696	15.29	15.32	15.26	0.1304	0.8817								
		8	0.2199	0.0063	0.0382	0.1996	0.8380	0.2204	0.0057	0.0388	0.1862	0.8157								
		9	0.1976	0.0078	0.0331	0.1996	0.8380	0.1970	0.0073	0.0320	0.1862	0.8157								

See Key

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																			
394	5	0.897	5.57	44.99	0.1375	15.34	15.30	15.33	15.26	0.1237	0.8796	0.2301	0.0063	0.0398	0.1950	0.8662	0.2314	0.0057	0.4077	0.1731	0.8141
	6	0.2032	0.0079	0.0337	0.0337	0.8303	0.2082	0.0072	0.0339	0.1731	0.8141	0.2032	0.0079	0.0337	0.0337	0.8303	0.2082	0.0072	0.0339	0.1731	0.8141
394	6	0.897	0.26	44.99	0.1619	15.30	15.28	15.30	15.24	0.1442	0.8867	0.1983	0.0064	0.0348	0.2079	0.8771	0.2012	0.0058	0.3566	0.2012	0.8217
	9	0.1825	0.0075	0.0311	0.0311	0.8533	0.1823	0.0073	0.0299	0.2012	0.8217	0.1825	0.0075	0.0311	0.0311	0.8533	0.1823	0.0073	0.0299	0.2012	0.8217
394	7	0.897	-0.76	44.99	0.1721	15.30	15.27	15.29	15.23	0.1553	0.8886	0.1865	0.0064	0.0328	0.2192	0.8809	0.1891	0.0058	0.3566	0.1553	0.8281
	9	0.1699	0.0074	0.0290	0.0290	0.8550	0.1743	0.0070	0.0288	0.1553	0.8281	0.1699	0.0074	0.0290	0.0290	0.8550	0.1743	0.0070	0.0288	0.1553	0.8281
395	3	0.600	-3.15	-0.00	0.2950	-0.02	-3.14	0.07	-3.12	0.3857	0.8797	0.0038	0.0002	0.0008	0.2373	1.1149	-0.0812	-0.0062	-0.0142	0.3857	0.8851
	9	0.0026	0.0001	0.0005	0.0005	0.9700	-0.0789	-0.0058	-0.0139	0.3857	0.8851	0.0026	0.0001	0.0005	0.0005	0.9700	-0.0789	-0.0058	-0.0139	0.3857	0.8851
395	4	0.600	-0.05	-0.00	0.3805	-0.00	-3.00	0.07	-3.01	0.3972	0.9347	0.0066	0.0005	0.0013	0.2776	0.9983	-0.0230	-0.0018	-0.0043	0.3972	0.9146
	9	0.0032	0.0001	0.0006	0.0006	0.9904	-0.0250	-0.0019	-0.0045	0.3972	0.9146	0.0032	0.0005	0.0013	0.0006	0.9904	-0.0250	-0.0019	-0.0045	0.3972	0.9146
395	5	0.600	0.99	-0.00	0.4011	-0.00	-2.96	0.06	-2.97	0.3878	1.0527	0.0064	0.0005	0.0012	0.2438	0.9653	-0.0068	-0.0007	-0.0020	0.3878	1.0527
	9	0.0035	0.0001	0.0006	0.0006	0.8719	-0.0103	-0.0007	-0.0020	0.3878	1.0527	0.0035	0.0005	0.0012	0.0006	0.8719	-0.0103	-0.0007	-0.0020	0.3878	1.0527
395	6	0.599	3.12	-0.00	0.4291	-0.01	-2.87	0.07	-2.90	0.3895	0.8439	0.0054	0.0004	0.0010	0.4291	0.9292	0.0242	0.0018	0.0034	0.3895	0.8702
	9	0.0038	0.0002	0.0006	0.0006	0.8878	0.0195	0.0016	0.0034	0.3895	0.8702	0.0038	0.0004	0.0010	0.0006	0.8878	0.0195	0.0016	0.0034	0.3895	0.8702
395	7	0.600	6.22	0.00	0.3535	-0.02	-2.74	0.08	-2.78	0.3872	0.8757	0.0033	0.0002	0.0009	0.3535	1.0048	0.0755	0.0058	0.0129	0.3872	0.8677
	9	0.0050	0.0003	0.0009	0.0009	0.9182	0.0746	0.0058	0.0129	0.3872	0.8677	0.0050	0.0003	0.0009	0.0009	0.9182	0.0746	0.0058	0.0129	0.3872	0.8677
395	8	0.601	9.35	0.00	0.2978	-0.02	-2.62	0.07	-2.67	0.3862	0.8256	0.0024	0.0001	0.0008	0.2427	1.0485	0.1221	0.0094	0.0199	0.3862	0.8069
	9	0.0046	0.0002	0.0008	0.0008	0.9762	0.1236	0.0092	0.0199	0.3862	0.8069	0.0046	0.0001	0.0008	0.0008	0.9762	0.1236	0.0092	0.0199	0.3862	0.8069
395	9	0.601	12.54	-0.00	0.3272	-0.01	-2.57	0.07	-2.62	0.3324	0.7801	0.0025	0.0001	0.0005	0.3272	1.0288	0.1597	0.0106	0.0245	0.3324	0.7674
	9	0.0024	0.0000	0.0005	0.0005	1.0907	0.1598	0.0102	0.0245	0.3324	0.7674	0.0025	0.0001	0.0005	0.0005	1.0288	0.1597	0.0106	0.0245	0.3324	0.7674
395	10	0.599	-0.07	-0.00	0.3731	-0.01	-3.00	0.07	-3.00	0.4129	0.9589	0.0063	0.0004	0.0012	0.3731	0.9878	-0.0229	-0.0018	-0.0043	0.4129	0.9406
	9	0.0030	0.0002	0.0006	0.0006	1.0311	-0.0252	-0.0019	-0.0047	0.4129	0.9406	0.0030	0.0004	0.0012	0.0006	1.0311	-0.0252	-0.0019	-0.0047	0.4129	0.9406

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key												
396	1	7	0.600	-0.12	-0.009	0.3522	-0.01	-0.18	0.07	-0.12	0.4079	0.8608			
	8	9	0.0044	0.0002	0.0006	0.3923	1.0972	-0.0428	-0.0036	-0.0079	0.3842	0.8784			
	9	9	0.0031	0.0002	0.0006	0.3923	1.1142	-0.0428	-0.0032	-0.0075	0.3842	0.8784			
396	2	7	0.600	-0.03	0.000	0.3958	-0.00	-0.05	0.07	-0.02	0.4220	0.9310			
	8	9	0.0065	0.0005	0.0012	0.2825	0.9786	0.0053	0.0004	0.0009	0.4585	0.9235			
	9	9	0.0032	0.0001	0.0006	0.2825	0.9451	0.0053	0.0003	0.0006	0.4585	0.9235			
396	3	7	0.600	1.04	-0.00	0.4206	-0.00	-0.01	0.07	0.01	0.3893	0.8902			
	8	9	0.0066	0.0005	0.0012	0.3184	0.9381	0.0229	0.0017	0.0040	0.4120	0.9260			
	9	9	0.0031	0.0001	0.0006	0.3184	1.0130	0.0180	0.0014	0.0033	0.4120	0.9260			
396	4	7	0.598	3.12	-0.00	0.4507	-0.01	0.07	0.07	0.09	0.3914	0.8877			
	8	9	0.0055	0.0005	0.0010	0.2755	0.9175	0.0584	0.0045	0.0103	0.3945	0.8897			
	9	9	0.0039	0.0002	0.0006	0.2755	0.8391	0.0534	0.0042	0.0095	0.3945	0.8897			
396	5	7	0.598	6.26	-0.00	0.3911	-0.01	0.19	0.08	0.21	0.3902	0.8547			
	8	9	0.0033	0.0002	0.0006	0.3417	1.0282	0.1067	0.0083	0.0182	0.3895	0.8566			
	9	9	0.0052	0.0003	0.0009	0.3417	0.8805	0.1063	0.0082	0.0177	0.3895	0.8566			
396	6	7	0.599	9.35	0.00	0.4256	-0.02	0.28	0.08	0.28	0.3634	0.7997			
	8	9	0.0025	0.0002	0.0005	0.2806	1.1506	0.1470	0.0106	0.0235	0.3536	0.7997			
	9	9	0.0049	0.0002	0.0009	0.2806	0.9994	0.1476	0.0104	0.0231	0.3536	0.7997			
396	7	7	0.599	12.56	-0.00	0.4672	-0.02	0.28	0.07	0.27	0.2768	0.7695			
	8	9	0.0022	0.0002	0.0005	0.2023	1.1628	0.1721	0.0095	0.0265	0.2594	0.7571			
	9	9	0.0028	0.0001	0.0005	0.2023	0.9640	0.1735	0.0090	0.0262	0.2594	0.7571			
396	8	7	0.599	-0.03	-0.00	0.4151	-0.00	-0.04	0.07	-0.02	0.3865	0.8475			
	8	9	0.0066	0.0005	0.0013	0.3527	0.9971	0.0054	0.0004	0.0009	0.4167	0.7166			
	9	9	0.0033	0.0002	0.0006	0.3527	1.0216	0.0034	0.0002	0.0004	0.4167	0.7166			
397	1	7	0.599	-3.10	-0.00	0.3456	-0.01	0.92	0.06	0.90	0.4177	0.8790			
	8	9	0.0042	0.0002	0.0009	0.3503	1.1228	-0.0317	-0.0026	-0.0055	0.3904	0.8672			
	9	9	0.0027	0.0001	0.0005	0.3503	1.0372	-0.0309	-0.0024	-0.0054	0.3904	0.8672			
397	2	7	0.600	-0.03	-0.00	0.3949	-0.01	1.07	0.07	1.00	0.3876	0.9193			
	8	9	0.0069	0.0005	0.0013	0.2971	0.9980	0.0169	0.0013	0.0031	0.4182	0.9437			
	9	9	0.0035	0.0002	0.0006	0.2971	0.9416	0.0134	0.0011	0.0025	0.4182	0.9437			
397	3	7	0.599	1.01	-0.00	0.4151	-0.00	1.12	0.07	1.03	0.3882	0.9261			
	8	9	0.0067	0.0005	0.0013	0.2509	0.9740	0.0346	0.0026	0.0062	0.4016	0.9261			
	9	9	0.0036	0.0001	0.0006	0.2509	0.8883	0.0296	0.0023	0.0055	0.4016	0.9261			

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																					
397	4	7	0.599	5.15	-0.00	0.4432	-0.01	0.718	0.07	1.11	0.3830	0.8929	0.0056	0.0005	0.0010	0.0006	0.2924	0.9438	0.0654	0.0055	0.0128	0.3902	0.8927	
		6	0.0037	0.0002	0.0006	0.0006	0.8658	0.0654	0.0051	0.0116														
397	5	7	0.600	6.27	-0.00	0.3803	-0.01	1.165	0.08	1.22	0.3846	0.8396	0.0034	0.0002	0.0007	0.0009	0.3769	1.0384	0.1156	0.0089	0.0195	0.3799	0.8261	
		6	0.0050	0.0003	0.0009	0.0009	0.9562	0.1156	0.0087	0.0191														
397	6	7	0.600	9.40	0.00	0.3901	-0.01	1.547	0.08	1.28	0.3496	0.7906	0.0036	0.0002	0.0007	0.0010	0.3066	1.0461	0.1537	0.0108	0.0244	0.3411	0.7779	
		6	0.0051	0.0003	0.0010	0.0010	1.0076	0.1537	0.0104	0.0239														
397	7	7	0.599	12.56	-0.00	0.3821	-0.01	1.761	0.07	1.25	0.2473	0.7718	0.0034	0.0002	0.0007	0.0006	0.2205	1.0643	0.1759	0.0087	0.0271	0.2398	0.7577	
		6	0.0034	0.0001	0.0006	0.0006	0.9721	0.1759	0.0084	0.0266														
397	8	7	0.600	-0.00	-0.00	0.3981	-0.00	1.08	0.06	0.99	0.3719	0.8847	0.0065	0.0005	0.0013	0.0006	0.3981	1.0123	0.108	0.0013	0.0031	0.4038	0.9072	
		6	0.0030	0.0001	0.0006	0.0006	1.0016	0.108	0.0011	0.0025														
398	1	7	0.600	-3.08	-0.00	0.3634	-0.01	2.95	0.07	3.03	0.4794	0.8973	0.0042	0.0003	0.0009	0.0006	0.3035	1.1677	-0.0087	-0.0006	-0.0017	0.4794	0.9072	
		6	0.0032	0.0001	0.0006	0.0006	0.9611	-0.0087	-0.0006	-0.0015														
398	2	7	0.600	0.00	-0.00	0.4111	-0.01	3.07	0.06	3.13	0.3829	0.9067	0.0062	0.0005	0.0012	0.0005	0.4111	1.0230	0.0396	0.0030	0.0071	0.3893	0.9105	
		6	0.0030	0.0001	0.0005	0.0005	0.9085	0.0375	0.0029	0.0068														
398	3	7	0.601	1.08	-0.00	0.4394	-0.00	3.12	0.07	3.18	0.3855	0.9053	0.0064	0.0005	0.0012	0.0006	0.4394	0.9814	0.0583	0.0045	0.0105	0.3874	0.9016	
		6	0.0035	0.0001	0.0006	0.0006	0.9256	0.0548	0.0042	0.0098														
398	4	7	0.600	3.19	-0.00	0.4487	-0.01	3.20	0.07	3.26	0.3819	0.8771	0.0054	0.0004	0.0009	0.0006	0.4487	0.9130	0.0941	0.0071	0.0165	0.3742	0.8843	
		6	0.0035	0.0002	0.0006	0.0006	0.9833	0.0915	0.0068	0.0161														
398	5	7	0.600	6.30	-0.00	0.3932	-0.01	3.30	0.08	3.35	0.3636	0.8247	0.0033	0.0002	0.0006	0.0009	0.3932	1.0390	0.1363	0.0099	0.0224	0.3579	0.8095	
		6	0.0031	0.0003	0.0009	0.0009	0.9344	0.1369	0.0098	0.0221														
398	6	7	0.600	9.44	0.00	0.3897	-0.02	3.34	0.07	3.38	0.3174	0.7747	0.0032	0.0002	0.0006	0.0010	0.3897	1.0471	0.1652	0.0104	0.0256	0.3022	0.7606	
		6	0.0030	0.0002	0.0010	0.0010	1.0037	0.1661	0.0100	0.0252														

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key	←															→														
398	7	7	0.601 0.0028 0.0028	12.57 0.0002 0.0001	-0.00 0.0005 0.0005	0.3908 0.1877	-0.02 1.0266 0.9409	3.29 0.1754 0.1753	0.07 0.0076 0.0072	3.33 0.0271 0.0266	0.2174 0.2071	0.7737 0.7605																					
398	8	7	0.600 0.0062 0.0031	-0.00 0.0005 0.0002	-0.00 0.0012 0.0006	0.4181 0.3155	-0.01 1.0074 1.0007	3.06 0.0394 0.0378	0.07 0.0029 0.0029	3.14 0.0070 0.0068	0.3800 0.3842	0.8945 0.8994																					
399	1	7	0.600 0.0039 0.0029	-3.02 0.0002 0.0001	-0.00 0.0008 0.0005	0.3487 0.2975	-0.01 1.145 0.9623	6.05 0.0204 0.0214	0.06 0.0014 0.0016	6.04 0.0039 0.0040	0.3605 0.3880	0.9664 0.9499																					
399	2	7	0.601 0.0066 0.0033	0.04 0.0005 0.0001	-0.00 0.0013 0.0006	0.4086 0.2633	-0.00 1.0040 0.9121	6.18 0.0753 0.0728	0.06 0.0056 0.0054	6.16 0.0137 0.0131	0.3728 0.3722	0.9123 0.9051																					
399	3	7	0.600 0.0070 0.0034	1.12 0.0005 0.0002	-0.00 0.0013 0.0006	0.4103 0.3046	-0.01 0.9421 1.0028	6.23 0.0937 0.0918	0.07 0.0070 0.0066	6.20 0.0166 0.0163	0.3775 0.3641	0.8884 0.8881																					
399	4	7	0.599 0.0054 0.0035	3.23 0.0005 0.0002	-0.00 0.0010 0.0006	0.4617 0.3546	-0.01 0.9399 0.9879	6.29 0.1231 0.1209	0.07 0.0091 0.0087	6.26 0.0207 0.0200	0.3708 0.3627	0.8406 0.8281																					
399	5	7	0.600 0.0029 0.0047	6.32 0.0002 0.0003	0.00 0.0006 0.0008	0.3935 0.3635	-0.02 1.0210 0.9242	6.36 0.1562 0.1567	0.07 0.0107 0.0104	6.32 0.0248 0.0243	0.3447 0.3341	0.7937 0.7753																					
399	6	7	0.600 0.0033 0.0049	9.44 0.0002 0.0003	0.00 0.0007 0.0010	0.4343 0.3135	-0.01 1.1158 1.0388	6.33 0.1746 0.1763	0.07 0.0086 0.0083	6.29 0.0270 0.0267	0.2474 0.2354	0.7734 0.7596																					
399	7	7	0.599 0.0032 0.0032	12.57 0.0002 0.0001	-0.00 0.0006 0.0006	0.3897 0.2408	-0.01 1.0471 1.0298	6.30 0.1775 0.1780	0.06 0.0067 0.0063	6.25 0.0275 0.0272	0.1904 0.1790	0.7757 0.7640																					
399	8	7	0.601 0.0063 0.0031	0.04 0.0005 0.0001	-0.00 0.0012 0.0006	0.4307 0.3118	-0.00 1.0103 1.0282	6.18 0.0761 0.0736	0.06 0.0056 0.0054	6.16 0.0137 0.0132	0.3707 0.3701	0.9041 0.8999																					
400	1	7	0.600 0.0043 0.0029	-3.00 0.0003 0.0001	-0.00 0.0009 0.0005	0.3640 0.3035	-0.01 1.1174 1.0224	9.16 0.0554 0.0554	0.06 0.0040 0.0041	9.10 0.0104 0.0102	0.3644 0.3734	0.9442 0.9273																					

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																			
400	2	7	0.600	0.0005	-0.000	0.4165	-0.001	0.1068	0.06	0.0185	0.3806	0.8697	0.0066	0.0001	0.0013	0.2600	1.0016	0.1052	0.0077	0.0179	0.3700	0.8535
400	3	7	0.600	1.14	-0.000	0.4468	-0.000	0.1219	0.06	0.207	0.3667	0.8491	0.0063	0.0001	0.0012	0.2951	0.9819	0.1201	0.0085	0.0201	0.3552	0.8366
400	4	7	0.600	3.27	-0.000	0.4599	-0.001	0.1495	0.07	0.28	0.3512	0.8111	0.0053	0.0002	0.0009	0.3207	0.9256	0.1484	0.0100	0.0237	0.3402	0.8004
400	5	7	0.600	6.34	-0.000	0.4129	-0.002	0.1692	0.07	0.29	0.2910	0.7747	0.0037	0.0002	0.0006	0.3865	0.9943	0.1704	0.0093	0.0259	0.2752	0.7604
400	6	7	0.600	9.42	0.000	0.4044	-0.001	0.1768	0.07	0.24	0.2056	0.7805	0.0030	0.0002	0.0009	0.2601	1.0975	0.1779	0.0069	0.0272	0.1941	0.7654
400	7	7	0.600	12.59	0.000	0.4457	-0.002	0.1764	0.06	0.20	0.1645	0.7833	0.0033	0.0001	0.0006	0.4408	1.0444	0.1764	0.0051	0.0273	0.1470	0.7736
400	8	7	0.600	0.08	-0.000	0.4317	-0.000	0.1077	0.07	0.21	0.3784	0.8653	0.0063	0.0002	0.0013	0.3244	1.0243	0.1059	0.0078	0.0180	0.3704	0.8534
401	1	7	0.600	-2.91	-0.000	0.3288	-0.001	0.1527	0.06	0.25	0.3533	0.8630	0.0048	0.0003	0.0010	0.3089	1.0578	0.1229	0.0084	0.0208	0.3440	0.8462
401	2	7	0.600	0.13	-0.000	0.4127	-0.000	0.1535	0.06	0.32	0.3356	0.8051	0.0064	0.0001	0.0006	0.2799	1.0028	0.1596	0.0103	0.0253	0.3230	0.7951
401	3	7	0.600	1.16	-0.000	0.4155	-0.000	0.1658	0.06	0.32	0.3179	0.7856	0.0069	0.0005	0.0012	0.2878	0.9362	0.1667	0.0103	0.0259	0.3117	0.7770
401	4	7	0.600	3.29	-0.000	0.4432	-0.001	0.1746	0.07	0.29	0.2488	0.7796	0.0056	0.0002	0.0010	0.4361	0.9438	0.1754	0.0086	0.0269	0.2405	0.7691

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key															
401	5	0.600	6.57	-0.00	0.4132	-0.01	15.50	0.07	15.26	0.1950	0.7831						
	6	0.0058	0.0005	0.0008	0.3515	1.0352	0.1789	0.0069	0.0280	0.1825	0.7732						
	9	0.0054	0.0003	0.0010	0.3515	0.9324	0.1904	0.0065	0.0279								
401	6	0.600	9.43	0.00	0.4074	-0.01	15.29	0.07	15.23	0.1650	0.7850						
	6	0.0038	0.0003	0.0007	0.2877	0.9933	0.1802	0.0059	0.0283	0.1455	0.7751						
	9	0.0052	0.0003	0.0010	0.2877	1.0108	0.1818	0.0052	0.0282								
401	7	0.600	12.65	-0.00	0.4252	-0.01	15.28	0.06	15.23	0.1492	0.7905						
	6	0.0032	0.0002	0.0006	0.2404	1.0462	0.1867	0.0055	0.0295	0.1341	0.7800						
	9	0.0028	0.0001	0.0005	0.2404	1.0639	0.1885	0.0050	0.0294								
401	8	0.598	0.15	-0.00	0.4323	-0.01	15.35	0.07	15.32	0.3345	0.8019						
	6	0.0065	0.0005	0.0013	0.3090	1.0056	0.1596	0.0106	0.0256	0.3226	0.7933						
	9	0.0032	0.0001	0.0006	0.3090	1.0157	0.1605	0.0103	0.0254								

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																			
403	3	7	1.251	-3.14	-0.00	0.4645	-0.01	-0.16	0.04	-0.10	0.1307	0.9516	0.0027	0.0002	0.0006	0.4645	1.2360	-0.0524	-0.0013	-0.0099	0.1307	0.9516
		8	0.0005	-0.0001	-0.0001	1.6347	1.4822	-0.0497	-0.0012	-0.0096	0.1207	0.9650	-0.0005	-0.0001	-0.0001	1.6347	1.4822	-0.0497	-0.0012	-0.0096	0.1207	0.9650
403	4	7	1.248	-0.01	-0.00	0.3458	-0.00	-0.06	0.05	-0.01	0.2289	1.0341	0.0047	0.0003	0.0009	0.3458	1.0426	0.0040	0.0001	0.0008	0.2289	1.0341
		8	0.0004	-0.0001	-0.0000	-1.8023	-0.5085	0.0031	0.0002	0.0005	0.3526	0.8595	0.0004	-0.0001	-0.0000	-1.8023	-0.5085	0.0031	0.0002	0.0005	0.3526	0.8595
403	5	7	1.249	1.05	0.00	0.3167	-0.01	-0.01	0.04	0.00	0.1554	0.9352	0.0051	0.0003	0.0009	0.3167	0.9563	0.0246	0.0007	0.0040	0.1554	0.9352
		8	0.0003	-0.0000	0.0000	-1.1865	0.4022	0.0217	0.0007	0.0040	0.1611	0.9448	0.0003	-0.0000	0.0000	-1.1865	0.4022	0.0217	0.0007	0.0040	0.1611	0.9352
403	6	7	1.250	5.19	0.00	0.3538	-0.00	0.04	0.04	0.05	0.1195	0.9518	0.0043	0.0003	0.0008	0.3538	0.9662	0.0615	0.0014	0.0117	0.1195	0.9518
		8	0.0005	-0.0000	0.0000	-0.0519	0.6338	0.0585	0.0011	0.0110	0.0999	0.9413	0.0005	-0.0000	0.0000	-0.0519	0.6338	0.0585	0.0011	0.0110	0.0999	0.9413
403	7	7	1.250	6.37	-0.00	0.3843	-0.01	0.13	0.05	0.12	0.1020	0.9513	0.0025	0.0001	0.0005	0.3843	1.0107	0.1138	0.0023	0.0216	0.1020	0.9513
		8	0.0022	0.0000	0.0000	0.0227	0.8312	0.1166	0.0020	0.0217	0.0861	0.9317	0.0022	0.0000	0.0000	0.0227	0.8312	0.1166	0.0020	0.0217	0.0861	0.9317
403	8	7	1.249	5.54	0.00	0.3140	-0.01	0.21	0.04	0.18	0.1007	0.9281	0.0023	0.0001	0.0004	0.3140	1.0111	0.1613	0.0032	0.0299	0.1007	0.9281
		8	0.0017	-0.0001	0.0000	-0.3342	1.0547	0.1641	0.0028	0.0298	0.0861	0.9092	0.0017	-0.0001	0.0000	-0.3342	1.0547	0.1641	0.0028	0.0298	0.0861	0.9092
403	9	7	1.249	-0.02	-0.00	0.3468	-0.00	-0.05	0.04	-0.02	0.1296	0.8582	0.0053	0.0010	0.0010	0.3468	1.0353	0.0037	0.0000	0.0006	0.1296	0.8582
		8	0.0002	-0.0000	0.0000	-0.4956	1.1686	0.0019	0.0001	0.0003	0.3546	0.7644	0.0002	-0.0000	0.0000	-0.4956	1.1686	0.0019	0.0001	0.0003	0.3546	0.7644
404	3	7	1.250	-4.06	-0.00	0.6687	-0.01	6.00	0.05	6.01	1.3045	2.8112	0.0022	0.0002	0.0006	0.6687	1.3948	0.0007	0.0002	0.0004	1.3045	2.8112
		8	0.0000	-0.0001	-0.0000	65.8429	13.3554	0.0027	0.0003	0.0007	0.6600	1.3791	0.0000	-0.0001	-0.0000	65.8429	13.3554	0.0027	0.0003	0.0007	0.6600	1.3791
404	4	7	1.251	-3.03	-0.00	0.4898	-0.01	6.03	0.05	6.03	0.2212	1.0606	0.0027	0.0002	0.0008	0.4898	1.3491	0.0182	0.0008	0.0040	0.2212	1.0606
		8	0.0027	0.0002	0.0000	89.5631	53.5162	0.0199	0.0008	0.0040	0.2160	1.0265	0.0027	0.0002	0.0008	89.5631	53.5162	0.0199	0.0008	0.0040	0.2160	1.0265
404	5	7	1.249	0.10	-0.00	0.3741	-0.01	6.13	0.05	6.11	0.1155	0.9908	0.0048	0.0003	0.0010	0.3741	1.0742	0.0775	0.0017	0.0150	0.1155	0.9908
		8	0.0005	-0.0001	0.0000	-1.3036	0.2826	0.0774	0.0015	0.0153	0.1019	0.9731	0.0005	-0.0001	0.0000	-1.3036	0.2826	0.0774	0.0015	0.0153	0.1019	0.9731
404	6	7	1.250	1.18	-0.00	0.3726	-0.01	6.16	0.05	6.13	0.1036	0.9626	0.0049	0.0003	0.0010	0.3726	1.0280	0.0988	0.0020	0.0187	0.1036	0.9626
		8	0.0001	-0.0000	0.0000	-1.1839	1.4896	0.0972	0.0016	0.0187	0.0866	0.9422	0.0001	-0.0000	0.0000	-1.1839	1.4896	0.0972	0.0016	0.0187	0.0866	0.9422

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key →																		
404	1	1.249	5.30	-0.00	0.3563	-0.01	6.21	0.05	6.16	0.1002	0.9540	0.0038	0.0000	0.0007	0.9226	0.1319	0.0026	0.0251	0.0793	0.9380
		0.0000	0.0000	0.0000	1.8105	3.6055	0.1314	0.0020	0.0246											
404	8	1.250	6.47	0.00	0.3869	-0.01	6.28	0.06	6.23	0.0934	0.9292	0.0025	0.0004	0.0005	1.0698	0.1754	0.0032	0.0326	0.0765	0.9048
		0.0017	0.0000	0.0000	0.1706	1.1583	0.1791	0.0027	0.0324											
404	9	1.249	0.09	-0.00	0.3641	-0.01	6.13	0.05	6.10	0.1098	0.9828	0.0050	0.0000	0.0010	1.0390	0.0780	0.0017	0.0153	0.0980	0.9645
		0.0002	0.0000	0.0000	0.8118	-1.3656	0.0774	0.0015	0.0149											
405	1	1.251	-3.91	-0.00	0.5172	-0.01	15.18	0.05	15.13	0.0969	0.9904	0.0026	0.0006	0.0020	1.2755	0.1079	0.0020	0.0213	0.0829	0.9702
		0.0005	0.0000	0.0000	0.0854	-0.5029	0.1107	0.0018	0.0214											
405	2	1.250	-2.85	-0.00	0.3755	-0.01	15.21	0.05	15.15	0.0930	0.9738	0.0033	0.0007	0.0023	1.1653	0.1270	0.0023	0.0247	0.0744	0.9552
		0.0009	0.0000	0.0000	0.2879	0.2783	0.1297	0.0019	0.0247											
405	3	1.250	-1.86	-0.00	0.3282	-0.01	15.24	0.05	15.17	0.0929	0.9617	0.0039	0.0008	0.0026	1.1047	0.1439	0.0026	0.0276	0.0724	0.9423
		0.0009	0.0000	0.0000	0.1902	0.2361	0.1462	0.0021	0.0275											
405	4	1.251	0.25	-0.00	0.3568	-0.00	15.29	0.05	15.20	0.0921	0.9357	0.0050	0.0010	0.0032	1.0400	0.1753	0.0032	0.0324	0.0712	0.9192
		0.0006	0.0000	0.0000	-0.0970	-0.3583	0.1766	0.0025	0.0324											
405	5	1.251	-0.80	-0.00	0.3363	-0.01	15.27	0.05	15.18	0.0926	0.9481	0.0043	0.0009	0.0029	1.1121	0.1601	0.0029	0.0303	0.0718	0.9289
		0.0009	0.0000	0.0000	0.2063	0.1805	0.1620	0.0023	0.0301											
405	6	1.251	0.24	-0.00	0.3738	-0.00	15.29	0.06	15.20	0.0921	0.9355	0.0049	0.0010	0.0032	1.0581	0.1753	0.0032	0.0324	0.0710	0.9191
		0.0009	0.0000	0.0000	-0.0863	0.0861	0.1765	0.0025	0.0324											
406	3	0.898	-3.11	-0.00	0.4708	-0.01	0.16	0.04	0.10	0.2103	0.9246	0.0032	0.0008	-0.0024	1.2486	-0.0582	-0.0024	-0.0107	0.2028	0.9324
		0.0004	0.0000	0.0000	2.8650	1.9625	-0.0558	-0.0022	-0.0104											
406	4	0.902	-0.02	-0.00	0.4922	-0.01	-0.05	0.05	0.02	0.5530	1.0897	0.0058	0.0012	-0.0054	1.0658	0.0038	0.0006	0.0011	0.5166	1.0150
		0.0007	0.0001	0.0000	-0.9508	-0.5574	0.0038	0.0003	0.0007											

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																						
406	5	7	0.900	1.03	-0.00	0.5817	-0.00	-0.00	-0.02	0.05	0.00	0.2875	0.9600	0.0052	0.0006	0.0012	0.5817	1.0753	0.0264	0.0015	0.0051	0.2311	0.9607	
		8	0.0012	-0.0001	0.0000	-0.5097	0.0000	0.1019	0.0236	0.0010	0.0015	0.0045												
406	6	7	0.900	3.13	0.00	0.5528	0.00	-0.01	0.04	0.05	0.07	0.2409	0.9363	0.0053	0.0005	0.0010	0.5528	0.9630	0.0707	0.0034	0.0132	0.2197	0.9304	
		8	0.0025	-0.0001	0.0002	-0.3773	0.0002	0.5746	0.0672	0.0029	0.0125													
406	7	7	0.897	6.26	0.00	0.4416	0.00	-0.02	0.13	0.06	0.15	0.2101	0.8866	0.0034	0.0000	0.0005	0.4416	1.0614	0.1300	0.0054	0.0231	0.2132	0.8642	
		8	0.0036	0.0000	0.0005	0.0976	0.0005	0.7970	0.1302	0.0055	0.0225													
406	8	7	0.899	9.39	0.00	0.2280	0.00	-0.01	0.20	0.05	0.20	0.1919	0.8611	0.0035	0.0001	0.0007	0.2280	1.0225	0.1776	0.0068	0.0297	0.1843	0.8360	
		8	0.0036	0.0000	0.0006	0.0243	0.0006	0.9326	0.1777	0.0065	0.0297													
406	9	7	0.899	12.58	0.00	0.2561	0.00	-0.01	0.23	0.05	0.23	0.1698	0.8329	0.0031	0.0001	0.0006	0.2561	1.0867	0.2133	0.0072	0.0355	0.1566	0.8133	
		8	0.0015	-0.0001	0.0002	-0.3882	0.0002	0.9086	0.2151	0.0067	0.0349													
406	10	7	0.899	-0.03	-0.00	0.4807	-0.00	-0.01	-0.06	0.05	-0.02	0.4698	0.9799	0.0058	0.0005	0.0012	0.4807	1.0501	0.0054	0.0005	0.0010	0.4187	0.8454	
		8	0.0004	-0.0000	-0.0000	-0.4358	-0.0000	-0.4982	0.0039	0.0003	0.0006													
407	1	7	0.902	-3.02	-0.00	0.3200	-0.00	-0.01	6.05	0.05	6.04	0.2953	1.0368	0.0041	0.0002	0.0008	0.3200	1.0445	0.0255	0.0015	0.0056	0.2399	1.0152	
		8	0.0002	-0.0001	-0.0001	3.8512	-0.0001	2.8193	0.0278	0.0013	0.0056													
407	2	7	0.899	-3.03	-0.00	0.3688	-0.00	-0.02	6.04	0.05	6.04	0.2770	1.0314	0.0041	0.0003	0.0009	0.3688	1.1081	0.0261	0.0014	0.0053	0.2342	1.0196	
		8	0.0002	-0.0001	-0.0001	4.3778	-0.0001	3.5682	0.0279	0.0013	0.0056													
407	3	7	0.899	0.07	0.00	0.4491	0.00	-0.01	6.15	0.05	6.13	0.2012	0.9470	0.0062	0.0005	0.0012	0.4491	1.0121	0.0944	0.0038	0.0178	0.1959	0.9227	
		8	0.0014	-0.0001	0.0000	-0.4275	0.0000	0.2458	0.0930	0.0036	0.0171													
407	4	7	0.898	1.14	0.00	0.4203	0.00	-0.01	6.18	0.05	6.16	0.2086	0.9148	0.0064	0.0005	0.0012	0.4203	0.9862	0.1134	0.0047	0.0207	0.2046	0.8975	
		8	0.0016	-0.0001	0.0001	-0.4711	0.0001	0.3418	0.1108	0.0045	0.0199													
407	5	7	0.900	2.16	0.00	0.4356	0.00	-0.01	6.21	0.05	6.17	0.2149	0.8907	0.0064	0.0005	0.0011	0.4356	0.9253	0.1283	0.0055	0.0228	0.2046	0.8751	
		8	0.0017	-0.0001	0.0003	-0.3628	0.0003	0.9294	0.1260	0.0051	0.0220													

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Pt.	Cd	See Key																		
407	6	7	0.898	5.17	0.0010	0.0003	0.3933	-0.2298	-0.01	0.882	0.7951	6.25	0.1443	0.1421	0.0058	0.0062	0.0058	0.2153	0.2072	0.8758	0.8600
		9	0.0059	0.0004	0.0000	0.0000	0.0000	0.0000	0.0000	0.882	0.7951	6.25	0.1443	0.1421	0.0058	0.0062	0.0058	0.2153	0.2072	0.8758	0.8600
		9	0.0021	-0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.7951	0.9240	0.1719	0.1733	0.0066	0.0069	0.0066	0.0066	0.2027	0.1913	0.8564	0.8352
407	7	7	0.898	5.32	0.0008	0.0004	0.3272	0.1788	-0.01	0.9516	0.9240	6.27	0.1719	0.1733	0.0066	0.0069	0.0066	0.2027	0.1913	0.8564	0.8352
		9	0.0045	0.0000	0.0000	0.0000	0.0000	0.0000	0.9516	0.9240	0.9240	0.1719	0.1733	0.0066	0.0069	0.0066	0.0066	0.2027	0.1913	0.8564	0.8352
		9	0.0026	0.0000	0.0000	0.0000	0.0000	0.0000	0.9240	0.9240	0.9240	0.1719	0.1733	0.0066	0.0069	0.0066	0.0066	0.2027	0.1913	0.8564	0.8352
407	8	7	0.898	6.33	0.0007	0.0005	0.3137	0.2504	-0.01	1.0556	0.9802	6.28	0.1840	0.1863	0.0070	0.0070	0.0067	0.1915	0.1815	0.8484	0.8272
		9	0.0037	0.0002	0.0000	0.0000	0.0000	0.0000	1.0556	0.9802	0.9802	0.1840	0.1863	0.0070	0.0070	0.0067	0.0067	0.1915	0.1815	0.8484	0.8272
		9	0.0029	0.0001	0.0000	0.0000	0.0000	0.0000	0.9802	0.9802	0.9802	0.1840	0.1863	0.0070	0.0070	0.0067	0.0067	0.1915	0.1815	0.8484	0.8272
407	9	7	0.899	8.44	0.0007	0.0007	0.2442	0.1966	-0.01	1.0625	1.0166	6.31	0.2052	0.2071	0.0072	0.0072	0.0067	0.1768	0.1637	0.8297	0.8085
		9	0.0037	0.0001	0.0000	0.0000	0.0000	0.0000	1.0625	1.0166	1.0166	0.2052	0.2071	0.0072	0.0072	0.0067	0.0067	0.1768	0.1637	0.8297	0.8085
		9	0.0034	0.0000	0.0000	0.0000	0.0000	0.0000	1.0166	1.0166	1.0166	0.2052	0.2071	0.0072	0.0072	0.0067	0.0067	0.1768	0.1637	0.8297	0.8085
407	10	7	0.899	9.45	0.0008	0.0006	0.1480	0.1030	-0.01	1.0217	1.0108	6.30	0.2147	0.2168	0.0069	0.0069	0.0063	0.1615	0.1463	0.8283	0.8071
		9	0.0040	0.0001	0.0000	0.0000	0.0000	0.0000	1.0217	1.0108	1.0108	0.2147	0.2168	0.0069	0.0069	0.0063	0.0063	0.1615	0.1463	0.8283	0.8071
		9	0.0030	0.0000	0.0000	0.0000	0.0000	0.0000	1.0108	1.0108	1.0108	0.2147	0.2168	0.0069	0.0069	0.0063	0.0063	0.1615	0.1463	0.8283	0.8071
407	11	7	0.899	12.59	0.0006	0.0003	0.2268	-0.1489	-0.01	0.9569	1.1252	6.29	0.2324	0.2339	0.0051	0.0051	0.0047	0.1110	0.1014	0.8253	0.8061
		9	0.0034	0.0001	0.0000	0.0000	0.0000	0.0000	0.9569	1.1252	1.1252	0.2324	0.2339	0.0051	0.0051	0.0047	0.0047	0.1110	0.1014	0.8253	0.8061
		9	0.0014	-0.0000	0.0000	0.0000	0.0000	0.0000	1.1252	1.1252	1.1252	0.2324	0.2339	0.0051	0.0051	0.0047	0.0047	0.1110	0.1014	0.8253	0.8061
407	12	7	0.900	0.08	-0.0000	-0.0001	0.4491	-0.3774	-0.01	1.0121	0.7431	6.15	0.0936	0.0923	0.0038	0.0038	0.0036	0.2042	0.1953	0.9350	0.9216
		9	0.0061	0.0005	0.0000	0.0000	0.0000	0.0000	1.0121	0.7431	0.7431	0.0936	0.0923	0.0038	0.0038	0.0036	0.0036	0.2042	0.1953	0.9350	0.9216
		9	0.0008	-0.0000	0.0000	0.0000	0.0000	0.0000	0.7431	0.7431	0.7431	0.0936	0.0923	0.0038	0.0038	0.0036	0.0036	0.2042	0.1953	0.9350	0.9216
408	1	7	0.901	-3.90	-0.0008	-0.0000	0.5356	-3.0977	-0.01	1.3021	-0.5183	15.20	0.1266	0.1288	0.0051	0.0051	0.0051	0.2028	0.1996	0.9040	0.8838
		9	0.0032	0.0003	0.0000	0.0000	0.0000	0.0000	1.3021	-0.5183	-0.5183	0.1266	0.1288	0.0051	0.0051	0.0051	0.0051	0.2028	0.1996	0.9040	0.8838
		9	0.0002	-0.0001	0.0000	0.0000	0.0000	0.0000	-0.5183	-0.5183	-0.5183	0.1266	0.1288	0.0051	0.0051	0.0051	0.0051	0.2028	0.1996	0.9040	0.8838
408	2	7	0.900	-2.88	-0.0001	-0.0000	0.4062	-4.8991	-0.01	1.138	-1.6092	15.23	0.1411	0.1438	0.0058	0.0058	0.0058	0.2028	0.2028	0.8892	0.8683
		9	0.0042	0.0003	0.0000	0.0000	0.0000	0.0000	1.138	-1.6092	-1.6092	0.1411	0.1438	0.0058	0.0058	0.0058	0.0058	0.2028	0.2028	0.8892	0.8683
		9	0.0001	-0.0001	0.0000	0.0000	0.0000	0.0000	-1.6092	-1.6092	-1.6092	0.1411	0.1438	0.0058	0.0058	0.0058	0.0058	0.2028	0.2028	0.8892	0.8683
408	3	7	0.901	0.17	-0.0014	-0.0001	0.3268	-0.4327	-0.01	0.9746	0.4842	15.29	0.1827	0.1843	0.0069	0.0069	0.0063	0.1905	0.1725	0.8545	0.8408
		9	0.0072	0.0004	0.0000	0.0000	0.0000	0.0000	0.9746	0.4842	0.4842	0.1827	0.1843	0.0069	0.0069	0.0063	0.0063	0.1905	0.1725	0.8545	0.8408
		9	0.0015	-0.0001	0.0000	0.0000	0.0000	0.0000	0.4842	0.4842	0.4842	0.1827	0.1843	0.0069	0.0069	0.0063	0.0063	0.1905	0.1725	0.8545	0.8408
408	4	7	0.901	1.21	-0.0013	-0.0002	0.3435	-0.3088	-0.01	0.9181	0.6263	15.29	0.1921	0.1937	0.0071	0.0071	0.0064	0.1855	0.1669	0.8412	0.8333
		9	0.0074	0.0005	0.0000	0.0000	0.0000	0.0000	0.9181	0.6263	0.6263	0.1921	0.1937	0.0071	0.0071	0.0064	0.0064	0.1855	0.1669	0.8412	0.8333
		9	0.0019	-0.0001	0.0000	0.0000	0.0000	0.0000	0.6263	0.6263	0.6263	0.1921	0.1937	0.0071	0.0071	0.0064	0.0064	0.1855	0.1669	0.8412	0.8333

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key															
408	5	7	0.898	2.25	-0.00	0.3567	-0.01	15.30	0.05	15.24	0.1791	0.8304						
		6	0.0067	0.0004	0.0012	0.2316	0.9405	0.2006	0.0071	0.0333	0.1644	0.8192						
		9	0.0016	-0.0000	0.0002	-0.2316	0.8730	0.2028	0.0066	0.0332								
408	6	7	0.898	3.30	0.00	0.3183	-0.01	15.30	0.05	15.24	0.1656	0.8271						
		6	0.0063	0.0004	0.0010	0.1280	0.8646	0.2090	0.0069	0.0345	0.1541	0.8127						
		9	0.0018	-0.0000	0.0002	-0.1280	0.8183	0.2090	0.0064	0.0339								
408	7	7	0.898	6.38	0.00	0.2761	-0.01	15.29	0.05	15.22	0.1218	0.8234						
		6	0.0047	0.0002	0.0009	0.2423	0.9788	0.2212	0.0053	0.0364	0.1100	0.8098						
		9	0.0030	0.0001	0.0005	0.2423	0.9042	0.2215	0.0048	0.0358								
408	8	7	0.900	9.46	0.00	0.2370	-0.01	15.29	0.06	15.23	0.1112	0.8211						
		6	0.0043	0.0002	0.0008	0.1230	1.0311	0.2325	0.0051	0.0381	0.0982	0.8066						
		9	0.0032	0.0000	0.0005	0.1230	0.8979	0.2338	0.0045	0.0377								
408	9	7	0.901	12.60	0.00	0.2735	-0.01	15.28	0.05	15.21	0.0997	0.8179						
		6	0.0043	0.0002	0.0007	0.0505	1.0328	0.2362	0.0047	0.0386	0.0828	0.8067						
		9	0.0015	0.0000	0.0003	0.0505	1.0328	0.2373	0.0039	0.0382								
408	10	7	0.900	0.14	-0.00	0.3418	-0.01	15.28	0.05	15.23	0.1906	0.8506						
		6	0.0068	0.0004	0.0013	0.3280	0.9756	0.1823	0.0069	0.0310	0.1730	0.8397						
		9	0.0013	-0.0000	0.0001	-0.3280	0.6850	0.1842	0.0063	0.0309								
409	3	7	0.598	-3.10	-0.00	0.3706	-0.02	-0.17	0.05	-0.12	0.4034	0.8758						
		6	0.0032	0.0002	0.0008	2.0266	1.2651	-0.0442	-0.0035	-0.0077	0.3883	0.8797						
		9	0.0003	-0.0001	-0.0000	-2.0266	-0.5102	-0.0425	-0.0033	-0.0074								
409	4	7	0.598	-0.04	-0.00	0.4040	-0.01	-0.04	0.05	-0.01	0.4512	1.0128						
		6	0.0059	0.0004	0.0012	0.4084	1.0320	0.0055	0.0005	0.0011	0.4383	0.9280						
		9	0.0013	-0.0001	0.0000	-0.4084	0.1006	0.0034	0.0003	0.0002								
409	5	7	0.599	1.03	0.00	0.4464	-0.00	-0.00	0.05	0.01	0.4025	0.9161						
		6	0.0062	0.0005	0.0012	0.3113	1.0005	0.0227	0.0016	0.0041	0.3934	0.9190						
		9	0.0016	-0.0001	0.0000	-0.3113	0.1675	0.0181	0.0014	0.0033								
409	6	7	0.598	5.13	0.00	0.4511	-0.01	0.09	0.05	0.09	0.3999	0.8982						
		6	0.0054	0.0004	0.0010	0.1794	0.9455	0.0584	0.0046	0.0105	0.3835	0.8911						
		9	0.0016	-0.0000	0.0001	-0.1794	0.3251	0.0536	0.0041	0.0095								
409	7	7	0.599	6.24	0.00	0.3865	-0.01	0.20	0.06	0.21	0.3943	0.8596						
		6	0.0036	0.0002	0.0007	0.1456	1.0118	0.1060	0.0083	0.0182	0.3864	0.8365						
		9	0.0029	0.0000	0.0003	0.1456	0.6016	0.1063	0.0082	0.0177								

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key →													
409	8	7	0.599	9.57	0.0002	0.0000	0.0007	0.4117	-0.01	0.1464	0.06	0.28	0.3661	0.8044		
	6	5	0.0031	0.0000	0.0000	0.0005	0.0851	1.1265	0.1472	0.0107	0.0235	0.3532	0.7856			
			0.0033	0.0000	0.0000	0.0005	0.0851	0.7771	0.1472	0.0104	0.0231					
409	9	7	0.599	12.52	0.0002	0.0006	0.3904	-0.01	0.29	0.05	0.27	0.2814	0.7709			
	6	5	0.0032	0.0000	0.0002	0.0002	0.1512	1.0777	0.1710	0.0096	0.0263	0.2641	0.7616			
			0.0018	-0.0000	0.0000	0.0002	-0.1512	0.5924	0.1723	0.0091	0.0262					
409	10	7	0.599	-0.02	0.0004	-0.0012	0.3644	-0.01	-0.04	0.05	-0.02	0.3852	0.9049			
	6	5	0.0064	0.0000	0.0004	0.0012	0.2577	0.9687	0.0059	0.0004	0.0010	0.3690	0.8607			
			0.0012	-0.0000	0.0000	0.0000	-0.2577	0.0900	0.0038	0.0002	0.0006					
410	1	7	0.599	-3.00	0.0002	-0.0009	0.3377	-0.01	6.04	0.05	6.04	0.3643	0.9853			
	6	5	0.0039	0.0002	0.0002	0.0009	0.5168	1.1652	0.0205	0.0014	0.0040	0.3933	0.9847			
			0.0010	-0.0001	-0.0000	-0.0000	-0.5168	-0.2298	0.0208	0.0016	0.0041					
410	2	7	0.599	0.04	0.0004	-0.0012	0.3724	-0.01	6.18	0.05	6.16	0.3744	0.9155			
	6	5	0.0063	0.0004	0.0004	0.0012	0.2183	0.9889	0.0751	0.0056	0.0137	0.3661	0.9011			
			0.0012	-0.0000	0.0000	0.0000	-0.2183	0.3046	0.0735	0.0053	0.0132					
410	3	7	0.598	1.11	0.0005	-0.0012	0.4290	-0.00	6.22	0.05	6.20	0.3807	0.8911			
	6	5	0.0064	0.0005	0.0005	0.0012	0.2867	0.9736	0.0935	0.0071	0.0166	0.3715	0.8789			
			0.0012	-0.0000	0.0000	0.0000	-0.2867	0.1638	0.0904	0.0067	0.0158					
410	4	7	0.599	5.21	0.0004	0.0009	0.4468	-0.01	6.28	0.05	6.26	0.3721	0.8436			
	6	5	0.0051	0.0004	0.0004	0.0009	0.1787	0.9372	0.1227	0.0091	0.0207	0.3637	0.8343			
			0.0014	-0.0000	0.0000	0.0000	-0.1787	0.2988	0.1196	0.0087	0.0199					
410	5	7	0.598	6.33	0.0002	0.0003	0.3926	-0.01	6.35	0.06	6.32	0.3449	0.7963			
	6	5	0.0038	0.0002	0.0001	0.0003	0.1839	1.0133	0.1566	0.0108	0.0249	0.3350	0.7787			
			0.0030	0.0001	0.0001	0.0003	0.1839	0.5978	0.1567	0.0105	0.0244					
410	6	7	0.599	9.42	0.0002	0.0007	0.3944	-0.01	6.32	0.06	6.29	0.2519	0.7757			
	6	5	0.0032	0.0002	0.0000	0.0007	0.0915	1.1161	0.1739	0.0087	0.0269	0.2358	0.7629			
			0.0034	0.0000	0.0000	0.0005	0.0915	0.7461	0.1757	0.0082	0.0268					
410	7	7	0.598	12.56	0.0002	0.0006	0.3786	-0.01	6.29	0.05	6.25	0.1919	0.7791			
	6	5	0.0030	0.0002	0.0002	0.0006	0.3207	1.0383	0.1774	0.0068	0.0276	0.1796	0.7670			
			0.0014	-0.0000	0.0000	0.0001	-0.3207	0.5074	0.1778	0.0063	0.0272					
410	8	7	0.599	0.05	0.0004	-0.0012	0.3976	-0.01	6.17	0.05	6.15	0.3714	0.9097			
	6	5	0.0060	0.0004	0.0004	0.0012	0.1383	1.0290	0.0758	0.0056	0.0137	0.3662	0.8996			
			0.0012	-0.0000	0.0000	0.0000	-0.1383	0.3686	0.0737	0.0054	0.0132					

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key															
411	6	7	0.599	-2.89	-0.00	0.3521	-0.01	15.28	0.05	15.26	0.3531	0.8650						
		6	0.0040	0.0002	0.0009	0.3521	1.1452	0.1211	0.0065	0.0209	0.3434	0.8650						
		9	0.0011	-0.0001	0.0000	-0.4561	0.0199	0.1230	0.0084	0.0208	0.3434	0.8650						
411	7	7	0.600	-0.90	-0.00	0.3952	-0.01	15.34	0.05	15.30	0.3425	0.8280						
		6	0.0057	0.0004	0.0012	0.3952	1.0790	0.1489	0.0102	0.0246	0.3290	0.8280						
		9	0.0011	-0.0000	0.0000	-0.2753	0.2736	0.1499	0.0098	0.0244	0.3290	0.8280						
411	8	7	0.600	0.16	-0.00	0.3919	-0.01	15.36	0.05	15.31	0.3365	0.8073						
		6	0.0064	0.0005	0.0012	0.3919	0.9701	0.1587	0.0106	0.0256	0.3240	0.8073						
		9	0.0014	-0.0000	0.0000	-0.2700	0.2152	0.1593	0.0103	0.0255	0.3240	0.8073						
411	9	7	0.600	1.19	-0.00	0.4534	-0.00	15.37	0.05	15.32	0.3200	0.7883						
		6	0.0062	0.0005	0.0012	0.4534	0.9682	0.1652	0.0105	0.0260	0.3139	0.7883						
		9	0.0011	-0.0000	0.0001	-0.2757	0.4629	0.1659	0.0104	0.0259	0.3139	0.7883						
411	10	7	0.600	3.28	-0.00	0.4617	-0.00	15.34	0.06	15.29	0.2498	0.7821						
		6	0.0054	0.0005	0.0010	0.4617	0.9399	0.1749	0.0087	0.0273	0.2463	0.7821						
		9	0.0016	-0.0000	0.0001	-0.0919	0.4164	0.1750	0.0086	0.0271	0.2463	0.7821						
411	11	7	0.600	6.35	0.00	0.4377	-0.01	15.31	0.06	15.26	0.1965	0.7857						
		6	0.0036	0.0003	0.0007	0.4377	1.0364	0.1785	0.0070	0.0280	0.1842	0.7857						
		9	0.0034	0.0001	0.0004	0.1543	0.5825	0.1798	0.0066	0.0279	0.1842	0.7857						
411	12	7	0.600	9.41	0.00	0.4340	-0.01	15.29	0.06	15.22	0.1654	0.7867						
		6	0.0035	0.0003	0.0007	0.4340	1.0336	0.1799	0.0059	0.0283	0.1474	0.7867						
		9	0.0033	0.0000	0.0004	0.0756	0.6817	0.1801	0.0053	0.0281	0.1474	0.7867						
411	13	7	0.600	12.60	0.00	0.4450	-0.01	15.29	0.05	15.23	0.1497	0.7931						
		6	0.0030	0.0002	0.0006	0.4450	1.1227	0.1864	0.0055	0.0295	0.1352	0.7931						
		9	0.0013	-0.0001	0.0001	-0.3801	0.3657	0.1874	0.0050	0.0293	0.1352	0.7931						
411	14	7	0.600	0.15	-0.00	0.3938	-0.00	15.36	0.05	15.32	0.3370	0.8062						
		6	0.0064	0.0005	0.0012	0.3938	0.9977	0.1587	0.0106	0.0255	0.3237	0.8062						
		9	0.0012	-0.0000	0.0000	-0.2542	0.2476	0.1593	0.0103	0.0254	0.3237	0.8062						
412	6	7	0.600	-0.00	-0.00	0.4044	-3.02	-0.05	-3.06	-0.01	0.3899	0.9184						
		6	0.0234	-0.0018	-0.0042	0.4044	0.9138	0.0062	0.0004	0.0011	0.3883	0.9184						
		9	-0.0291	-0.0025	-0.0054	0.4394	0.9366	0.0041	0.0003	0.0007	0.3883	0.9184						
412	7	7	0.599	0.00	-0.00	0.4089	-2.05	-0.05	-2.08	-0.02	0.3781	0.9152						
		6	0.0127	-0.0010	-0.0022	0.4089	0.8764	0.0065	0.0004	0.0012	0.3707	0.9152						
		9	-0.0199	-0.0017	-0.0037	0.4461	0.9307	0.0040	0.0003	0.0006	0.3707	0.9152						

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																		
412	8	7	8	9	0.599	0.0002	0.0000	-0.0004	0.4157	0.4721	-1.04	0.7137	0.9478	-0.063	0.0041	-1.05	0.0004	0.0011	0.3790	0.8352
		8			-0.0093	-0.0008	-0.0017	-0.0004	0.4721	0.4721	0.7137	0.9478	0.9478	0.0041	0.0041	0.0002	0.0002	0.0006	0.3557	0.8381
412	9	7	8	9	0.600	0.0001	0.0000	0.0004	0.3794	0.5865	-0.53	1.2505	1.0784	-0.065	0.0040	-0.50	0.0004	0.0011	0.3786	0.9101
		8			-0.0044	-0.0005	-0.0009	0.0004	0.5865	0.5865	1.2505	1.0784	1.0784	0.0040	0.0040	0.0003	0.0003	0.0006	0.3877	0.8513
412	10	7	8	9	0.600	0.0001	0.0000	0.0004	0.3876	0.2094	-0.01	0.9774	0.4760	-0.062	0.0041	0.05	0.0004	0.0011	0.3808	0.9135
		8			0.0062	0.0004	0.0012	0.0012	0.2094	-0.2094	0.9774	0.4760	0.4760	0.0041	0.0041	0.0002	0.0002	0.0006	0.3383	0.7864
412	11	7	8	9	0.600	0.0009	0.0003	0.0022	0.3931	0.3100	0.52	0.9563	0.8851	-0.064	0.0040	0.54	0.0004	0.0011	0.3785	0.9191
		8			0.0056	0.0003	0.0009	0.0022	0.3100	0.3100	0.9563	0.8851	0.8851	0.0040	0.0040	0.0002	0.0002	0.0006	0.3366	0.8110
412	12	7	8	9	0.599	0.0013	0.0006	0.0032	0.4041	0.3378	1.05	0.9513	0.8797	-0.066	0.0042	1.03	0.0005	0.0012	0.3772	0.9144
		8			0.0171	0.0013	0.0006	0.0032	0.3378	0.3378	0.9513	0.8797	0.8797	0.0042	0.0042	0.0002	0.0002	0.0006	0.3246	0.8158
412	13	7	8	9	0.600	0.0022	0.0014	0.0052	0.3969	0.3568	2.08	0.9255	0.9051	-0.068	0.0043	2.02	0.0005	0.0012	0.3690	0.9092
		8			0.0283	0.0022	0.0014	0.0052	0.3568	0.3568	0.9255	0.9051	0.9051	0.0043	0.0043	0.0002	0.0002	0.0007	0.3433	0.8606
412	14	7	8	9	0.599	0.0032	0.0022	0.0075	0.3951	0.3679	3.11	0.9259	0.9171	-0.069	0.0042	2.99	0.0005	0.0012	0.3749	0.9177
		8			0.0405	0.0032	0.0022	0.0075	0.3679	0.3679	0.9259	0.9171	0.9171	0.0042	0.0042	0.0002	0.0002	0.0007	0.3424	0.8444
412	15	7	8	9	0.599	0.0004	0.0000	0.0012	0.3615	0.0516	-0.02	0.9579	0.7523	-0.066	0.0041	0.06	0.0005	0.0012	0.3773	0.9163
		8			0.0063	0.0004	0.0000	0.0012	0.0516	-0.0516	0.9579	0.7523	0.7523	0.0041	0.0041	0.0003	0.0003	0.0007	0.3720	0.8630
413	3	7	8	9	1.250	0.0017	0.0001	0.0003	-0.2919	1.0332	-0.02	1.0332	1.7896	-0.265	-0.0228	0.05	0.0005	-0.0036	0.0943	0.6931
		8			0.0007	0.0001	0.0001	0.0002	-1.0165	1.7896	1.0332	1.7896	1.7896	-0.0228	-0.0228	-0.0005	-0.0005	0.0030	0.1268	0.6783
413	4	7	8	9	1.252	0.0020	0.0001	0.0004	-0.1815	7.0262	-0.02	0.9821	7.0262	-0.064	-0.0048	0.06	0.0002	-0.0009	0.1625	0.7489
		8			0.0001	0.0001	0.0001	0.0002	-3.4598	7.0262	0.02	0.9821	7.0262	-0.0048	-0.0048	-0.0002	-0.0002	0.0006	0.2892	0.6715

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key										
413	5	7	1.252	-0.0000	-0.0000	-0.0000	0.0000	-0.07	0.06	-0.02	0.0507	0.6334
		8	0.0026	-0.0000	-0.0004	0.0004	-0.0013	0.0013	0.0000	0.0001	0.0004	0.0000
		9	-0.0001	0.0001	-0.0002	-0.0002	0.0163	0.0028	-0.0001	0.0000	-0.02017	0.6204
413	6	7	1.252	0.50	-0.00	-0.00	-0.02	-0.06	0.05	-0.02	0.1563	0.7572
		8	0.0027	-0.0000	0.0004	0.0004	0.8780	0.0049	0.0001	0.0007	0.0007	0.0000
		9	-0.0001	0.0001	-0.0001	-0.0001	7.9974	0.0062	-0.0000	0.0009	-0.0071	0.7537
413	7	7	1.252	1.02	-0.00	-0.00	-0.02	-0.04	0.06	-0.01	0.1156	0.6974
		8	0.0030	-0.0000	0.0004	0.0004	0.8161	0.0098	0.0002	0.0013	0.0014	0.0000
		9	-0.0001	0.0001	-0.0001	-0.0001	9.7443	0.0101	0.0000	0.0014	0.0462	0.7214
413	8	7	1.251	3.12	-0.00	-0.00	-0.02	-0.02	0.06	0.00	0.0845	0.6720
		8	0.0028	-0.0000	0.0004	0.0004	0.7704	0.0280	0.0004	0.0037	0.0038	0.0000
		9	0.0000	0.0002	-0.0001	-0.0001	10.4414	0.0278	0.0003	0.0038	0.0038	0.6854
413	9	7	1.251	6.26	-0.00	-0.00	-0.02	0.01	0.06	0.03	0.0683	0.6746
		8	0.0019	-0.0001	0.0003	0.0003	0.8562	0.0547	0.0007	0.0073	0.0077	0.0000
		9	0.0007	0.0002	-0.0001	-0.0001	-0.7355	0.0575	0.0004	0.0077	0.0350	0.6741
413	10	7	1.251	9.39	-0.00	-0.00	-0.02	0.05	0.06	0.05	0.0645	0.6542
		8	0.0020	-0.0001	0.0003	0.0003	0.7579	0.0804	0.0010	0.0105	0.0109	0.0000
		9	0.0000	0.0002	-0.0000	-0.0000	9.8492	0.0842	0.0006	0.0109	0.0364	0.6525
413	11	7	1.251	12.62	-0.00	-0.00	-0.02	0.08	0.06	0.08	0.0624	0.6330
		8	0.0016	-0.0000	0.0002	0.0002	0.7972	0.1014	0.0012	0.0128	0.0133	0.0000
		9	-0.0006	0.0002	-0.0001	-0.0001	1.4611	0.1053	0.0008	0.0133	0.0408	0.6329
413	12	7	1.253	0.02	-0.00	-0.00	-0.02	-0.07	0.06	-0.02	-0.1309	0.4090
		8	0.0027	-0.0000	0.0004	0.0004	0.8030	0.0014	0.0000	0.0001	0.0003	0.0000
		9	-0.0003	0.0002	-0.0002	-0.0002	2.6515	0.0026	-0.0001	0.0003	-0.2555	0.6672
414	1	7	1.252	3.05	-0.00	-0.00	-0.02	6.00	0.05	6.00	0.0912	0.7537
		8	0.0019	-0.0001	0.0003	0.0003	0.816	0.0069	0.0001	0.0010	0.0172	0.7442
		9	-0.0007	0.0001	-0.0002	-0.0002	1.9100	0.0086	0.0000	0.0012	0.0172	0.7442
414	2	7	1.251	0.98	-0.00	-0.00	-0.02	6.03	0.06	6.03	0.0822	0.6931
		8	0.0024	-0.0000	0.0004	0.0004	0.8133	0.0262	0.0004	0.0036	0.0544	0.6931
		9	-0.0004	0.0001	-0.0002	-0.0002	2.7149	0.0271	0.0002	0.0038	0.0544	0.7021
9414	3	7	1.251	0.04	-0.00	-0.00	-0.02	6.05	0.06	6.05	0.0745	0.6907
		8	0.0029	-0.0000	0.0004	0.0004	0.7798	0.0361	0.0005	0.0049	0.0745	0.6907
		9	-0.0003	0.0002	-0.0002	-0.0002	2.7375	0.0363	0.0003	0.0050	0.0498	0.6962

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																	
414	4	7 8 9	1.251 0.0030 -0.0003	0.0000 0.0002 -0.0001	-0.0004 -0.1263 -3.2365	-0.02 0.7980 2.9202	6.05 0.0411 0.0413	0.06 0.0005 0.0003	6.04 0.0056 0.0056	0.0716 0.0469 0.0469	0.6904 0.6884 0.6884									
414	5	7 8 9	1.252 0.0031 -0.0002	0.0000 0.0005 -0.0001	-0.1127 -4.3050	-0.02 0.8256 3.7655	6.06 0.0461 0.0453	0.06 0.0006 0.0004	6.05 0.0063 0.0063	0.0687 0.0469 0.0469	0.6862 0.6913 0.6913									
414	6	7 8 9	1.251 0.0030 -0.0002	0.0000 0.0004 -0.0001	-0.1603 -5.0812	-0.02 0.7656 3.7602	6.09 0.0633 0.0638	0.06 0.0008 0.0005	6.06 0.0085 0.0086	0.0670 0.0403 0.0403	0.6744 0.6787 0.6787									
414	7	7 8 9	1.252 0.0021 0.0004	0.0003 0.0000 -0.0000	-0.2822 -2.9820	-0.02 0.3108 -0.9963	6.12 0.0865 0.0900	0.06 0.0010 0.0006	6.09 0.0112 0.0116	0.0600 0.0372 0.0372	0.6521 0.6487 0.6487									
414	8	7 8 9	1.252 0.0022 -0.0000	0.0003 0.0001 -0.0002	-0.2652 -0.4790	-0.02 0.7401 1.1653	6.15 0.1067 0.1107	0.06 0.0012 0.0008	6.12 0.0134 0.0139	0.0589 0.0388 0.0388	0.6315 0.6280 0.6280									
414	9	7 8 9	1.251 0.0017 -0.0007	0.0000 0.0002 -0.0002	-0.2738 -1.9456	-0.02 0.7561 1.4782	6.17 0.1225 0.1243	0.06 0.0013 0.0011	6.14 0.0150 0.0150	0.0570 0.0457 0.0457	0.6152 0.6050 0.6050									
414	10	7 8 9	1.252 0.0028 -0.0004	0.0000 0.0002 -0.0002	-0.1348 -2.7212	-0.02 0.7920 2.2596	6.05 0.0360 0.0362	0.06 0.0005 0.0003	6.04 0.0049 0.0050	0.0728 0.0494 0.0494	0.6876 0.6946 0.6946									
415	1	7 8 9	1.253 0.0019 -0.0006	0.0001 0.0003 -0.0003	-0.3026 -1.4353	-0.02 0.8650 2.2476	15.09 0.0589 0.0613	0.05 0.0007 0.0004	15.05 0.0081 0.0084	0.0649 0.0404 0.0404	0.6927 0.6925 0.6925									
415	2	7 8 9	1.252 0.0023 -0.0006	0.0000 0.0002 -0.0002	-0.1702 -1.6180	-0.02 0.8852 2.2933	15.12 0.0767 0.0785	0.06 0.0010 0.0006	15.07 0.0102 0.0105	0.0662 0.0411 0.0411	0.6708 0.6725 0.6725									
415	3	7 8 9	1.253 0.0029 -0.0003	0.0000 0.0002 -0.0002	-0.1295 -2.9735	-0.02 0.7798 4.1072	15.13 0.0844 0.0863	0.06 0.0011 0.0006	15.08 0.0111 0.0114	0.0662 0.0378 0.0378	0.6618 0.6606 0.6606									
415	4	7 8 9	1.251 0.0030 -0.0003	0.0000 0.0002 -0.0002	-0.1101 -2.8264	-0.02 0.8086 3.6076	15.13 0.0884 0.0900	0.06 0.0011 0.0006	15.08 0.0116 0.0118	0.0652 0.0379 0.0379	0.6562 0.6572 0.6572									

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key															
415	5	7	1.251	1.16	-0.000	-0.000	-0.000	-0.000	-0.104	-0.02	15.14	0.06	15.09	0.0646	0.6533		
		8	0.0031	-0.000	0.000	0.000	0.000	0.000	0.1047	0.8195	0.0919	0.0011	0.0120	0.0383	0.6543		
		9	-0.0004	0.000	-0.000	-0.000	-0.000	-2.3272	2.5616	0.0934	0.0007	0.0122					
415	6	7	1.253	5.27	-0.000	-0.000	-0.000	-0.1446	-0.02	15.16	0.06	15.10	0.0599	0.6348			
		8	0.0030	-0.000	0.000	0.000	0.000	0.1446	0.7318	0.1051	0.0012	0.0133	0.0378	0.6370			
		9	-0.0005	0.000	-0.000	-0.000	-2.7776	2.1780	0.1070	0.0008	0.0136						
415	7	7	1.247	6.39	-0.000	-0.000	-0.2655	-0.02	15.18	0.05	15.12	0.0623	0.6174				
		8	0.0023	-0.000	0.000	0.000	0.2655	0.7348	0.1202	0.0014	0.0148	0.0509	0.6085				
		9	0.0005	0.000	-0.000	-2.3536	-0.6909	0.1222	0.0012	0.0148							
415	8	7	1.251	9.50	-0.000	-0.000	-0.2956	-0.02	15.19	0.06	15.14	0.0573	0.5982				
		8	0.0022	-0.000	0.000	0.000	0.2956	0.7401	0.1303	0.0014	0.0156	0.0373	0.6012				
		9	-0.0001	0.000	-0.000	-9.4244	5.3045	0.1351	0.0010	0.0161							
415	9	7	1.252	12.70	-0.000	-0.000	-1.4990	-0.02	15.20	0.06	15.14	0.0511	0.5862				
		8	0.0018	-0.000	0.000	0.000	1.4990	0.7797	0.1378	0.0014	0.0163	0.0353	0.5935				
		9	-0.0009	0.000	-0.000	-1.9992	1.2578	0.1407	0.0009	0.0164							
415	10	7	1.251	0.09	-0.000	-0.000	-0.1263	-0.02	15.13	0.06	15.08	0.0643	0.6580				
		8	0.0028	-0.000	0.000	0.000	0.1263	0.7849	0.0844	0.0010	0.0111	0.0380	0.6598				
		9	-0.0006	0.000	-0.000	-1.9992	2.2375	0.0860	0.0006	0.0113							
417	1	7	1.252	-3.15	-0.000	-0.000	-0.2793	-0.02	3.00	0.05	-3.00	0.0743	0.6887				
		8	0.0019	-0.000	0.000	0.000	0.2793	0.9826	-0.0442	-0.0006	-0.0060	0.0827	0.6915				
		9	-0.0012	0.000	-0.000	-0.7652	0.8219	-0.0406	-0.0006	-0.0056							
417	2	7	1.249	-1.08	-0.000	-0.000	-0.1814	-0.02	-2.97	0.06	-2.98	0.0909	0.6979				
		8	0.0022	-0.000	0.000	0.000	0.1814	0.9439	-0.0237	-0.0004	-0.0033	0.1261	0.7124				
		9	-0.0007	0.000	-0.000	-1.2307	1.0303	-0.0207	-0.0005	-0.0029							
417	3	7	1.251	-0.04	-0.000	-0.000	-1.5372	-0.02	-2.96	0.06	-2.97	0.1287	0.7198				
		8	0.0029	-0.000	0.000	0.000	1.5372	0.8680	-0.0139	-0.0003	-0.0020	0.1656	0.7333				
		9	-0.0006	0.000	-0.000	-1.9481	1.1533	-0.0119	-0.0004	-0.0017							
417	4	7	1.251	0.47	-0.000	-0.000	-0.1073	-0.02	-2.95	0.06	-2.97	0.1646	0.7438				
		8	0.0031	-0.000	0.000	0.000	0.1073	0.8363	-0.0092	-0.0003	-0.0013	0.2340	0.7560				
		9	-0.0005	0.000	-0.000	-1.9481	1.2932	-0.0079	-0.0003	-0.0012							
417	5	7	1.251	1.00	-0.000	-0.000	-0.1020	-0.02	-2.95	0.06	-2.96	0.2608	0.7728				
		8	0.0031	-0.000	0.000	0.000	0.1020	0.8467	-0.0043	-0.0002	-0.0007	0.3416	0.7728				
		9	-0.0006	0.000	-0.000	-1.8654	0.9940	-0.0041	-0.0002	-0.0006							

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key														
417	6	1.250	5.12	-0.005	-0.1219	-0.02	-2.92	0.06	-2.94	0.0999	0.6590					
	6	0.0031	-0.0000	0.0005	-0.3763	0.7876	0.0105	0.0002	0.0013	0.0231	0.6567					
	9	-0.0003	0.0002	-0.0001	-5.3763	1.7756	0.0104	0.0000	0.0013	0.0231	0.6567					
417	7	1.251	6.23	-0.003	-0.2916	-0.02	-2.88	0.06	-2.91	0.0741	0.6711					
	8	0.0020	-0.0001	0.0003	-0.7105	0.8281	0.0380	0.0005	0.0051	0.0422	0.6663					
	9	0.0001	0.0002	-0.0000	11.7105	-1.6491	0.0397	0.0003	0.0052	0.0422	0.6663					
417	8	1.252	9.35	-0.003	-0.2938	-0.01	-2.84	0.06	-2.88	0.0669	0.6602					
	8	0.0019	-0.0001	0.0003	-2.4050	0.8008	0.0656	0.0008	0.0086	0.0387	0.6571					
	9	-0.0006	0.0002	-0.0000	-2.4050	0.5348	0.0687	0.0005	0.0090	0.0387	0.6571					
417	9	1.250	12.60	-0.002	-0.3222	-0.02	-2.80	0.06	-2.86	0.0651	0.6435					
	8	0.0016	-0.0001	0.0002	-1.3193	0.7981	0.0892	0.0011	0.0114	0.0419	0.6372					
	9	-0.0012	0.0003	-0.0001	-1.3193	0.6444	0.0930	0.0007	0.0118	0.0419	0.6372					
417	10	1.250	-0.07	-0.005	-0.1091	-0.01	-2.96	0.06	-2.97	0.1394	0.7413					
	8	0.0030	-0.0000	0.0005	-1.5459	0.8427	-0.0142	-0.0003	-0.0021	0.1766	0.7485					
	9	-0.0008	0.0002	-0.0001	-1.5459	0.6592	-0.0127	-0.0004	-0.0019	0.1766	0.7485					
418	1	1.251	-1.11	-0.003	-0.2704	-0.01	-1.01	0.06	-1.04	0.0886	0.6951					
	8	0.0020	-0.0001	0.0003	-0.8871	0.9311	-0.0320	-0.0005	-0.0044	0.1070	0.5968					
	9	-0.0014	0.0002	-0.0001	-0.8871	0.6581	-0.0292	-0.0006	-0.0040	0.1070	0.5968					
418	2	1.252	-1.05	-0.004	-0.1550	-0.01	-0.97	0.06	-1.01	0.1370	0.7090					
	8	0.0025	-0.0000	0.0004	-1.4114	0.8653	-0.0117	-0.0003	-0.0016	0.2020	0.7375					
	9	-0.0008	0.0002	-0.0001	-1.4114	0.9650	-0.0101	-0.0004	-0.0014	0.2020	0.7375					
418	3	1.251	-0.03	-0.005	-0.0980	-0.01	-0.96	0.06	-1.00	0.3069	0.8480					
	8	0.0029	-0.0000	0.0005	-1.6576	0.8741	-0.0027	-0.0001	-0.0004	0.5386	0.8605					
	9	-0.0007	0.0002	-0.0001	-1.6576	0.9458	-0.0024	-0.0002	-0.0004	0.5386	0.8605					
418	4	1.248	0.50	-0.005	-0.1139	-0.01	-0.95	0.06	-1.00	-0.1686	0.5038					
	8	0.0030	-0.0000	0.0005	-1.9513	0.8549	0.0013	-0.0000	0.0001	-0.4984	0.4574					
	9	-0.0006	0.0002	-0.0001	-1.9513	0.9837	0.0016	-0.0001	0.0001	-0.4984	0.4574					
418	5	1.250	0.98	-0.005	-0.1175	-0.01	-0.94	0.06	-1.00	0.0978	0.7220					
	8	0.0031	-0.0000	0.0005	-2.1682	0.8374	0.0044	0.0000	0.0006	-0.0549	0.6524					
	9	-0.0005	0.0002	-0.0001	-2.1682	1.1829	0.0045	-0.0000	0.0005	-0.0549	0.6524					
418	6	1.251	3.12	-0.004	-0.1409	-0.01	-0.92	0.06	-0.97	0.0862	0.6626					
	8	0.0032	-0.0000	0.0004	-2.2851	0.7613	0.0227	0.0003	0.0030	0.0554	0.6615					
	9	-0.0006	0.0002	-0.0001	-2.2851	1.2051	0.0212	0.0002	0.0028	0.0554	0.6615					

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key													
418	7	1.251	6.25	-0.0003	-0.0003	-0.2653	0.7592	-0.01	-0.88	0.06	-0.94	0.0671	0.6697		
	8	0.0023	-0.0001	0.0001	0.0003	-0.2653	0.7592	0.0500	0.0512	0.0006	0.0067	0.0370	0.6676		
	9	-0.0001	0.0003	-0.0000	-0.0000	-11.2702	0.1532	0.0512	0.0512	0.0003	0.0068	0.0370	0.6676		
418	7	1.251	9.37	-0.0001	-0.0003	-0.2637	0.7749	-0.02	-0.84	0.06	-0.91	0.0634	0.6541		
	8	0.0020	-0.0001	0.0001	0.0003	-0.2637	0.7749	0.0761	0.0791	0.0009	0.0099	0.0369	0.6475		
	9	-0.0005	0.0003	-0.0000	-0.0000	-3.1341	0.5665	0.0791	0.0791	0.0005	0.0102	0.0369	0.6475		
418	7	1.251	12.62	-0.0001	-0.0002	-0.3215	0.8411	-0.01	-0.80	0.06	-0.89	0.0629	0.6359		
	8	0.0016	-0.0001	0.0001	0.0002	-0.3215	0.8411	0.0979	0.1014	0.0012	0.0124	0.0409	0.6325		
	9	-0.0013	0.0003	-0.0001	-0.0001	-1.2335	0.6433	0.1014	0.1014	0.0008	0.0128	0.0409	0.6325		
418	7	1.250	-0.0004	-0.0005	-0.0001	-0.1109	0.8496	-0.01	-0.96	0.06	-1.01	0.3786	0.9788		
	8	0.0030	-0.0000	0.0005	0.0005	-0.1109	0.8496	0.0227	0.0227	0.0002	0.0005	0.4893	0.9277		
	9	-0.0008	0.0002	-0.0001	-0.0001	-1.5967	0.7780	-0.0227	-0.0227	-0.0002	-0.0005	0.4893	0.9277		
419	7	1.251	-3.10	-0.0001	-0.0003	-0.2703	0.9367	-0.02	-0.97	0.06	-0.91	0.1224	0.7055		
	8	0.0020	-0.0001	0.0003	0.0003	-0.2703	0.9367	0.0194	0.0194	0.0004	0.0027	0.1571	0.7035		
	9	-0.0013	0.0002	-0.0002	-0.0002	-1.0069	0.7530	-0.0173	-0.0173	-0.0005	-0.0024	0.1571	0.7035		
419	7	1.251	-1.05	-0.0002	-0.0004	-0.1459	0.8300	-0.01	-1.02	0.06	-0.94	0.7852	1.1082		
	8	0.0027	-0.0000	0.0004	0.0004	-0.1459	0.8300	0.0007	0.0007	0.0001	0.0001	5.0515	2.2058		
	9	-0.0008	0.0002	-0.0001	-0.0001	-1.4944	1.1276	-0.0001	-0.0001	-0.0001	-0.0000	5.0515	2.2058		
419	7	1.250	-0.0002	-0.0005	-0.0001	-0.1105	0.8613	-0.01	-1.03	0.06	-0.95	0.1073	0.7168		
	8	0.0030	-0.0000	0.0006	0.0005	-0.1105	0.8613	0.0065	0.0065	0.0001	0.0009	0.0176	0.6992		
	9	-0.0008	0.0002	-0.0001	-0.0001	-1.5447	0.8952	0.0065	0.0065	0.0000	0.0009	0.0176	0.6992		
419	7	1.249	0.52	-0.0002	-0.0005	-0.1130	0.8145	-0.01	-1.04	0.06	-0.96	0.0998	0.6922		
	8	0.0031	-0.0000	0.0006	0.0005	-0.1130	0.8145	0.0113	0.0113	0.0002	0.0015	0.0556	0.7060		
	9	-0.0006	0.0002	-0.0001	-0.0001	-2.0163	1.0036	0.0104	0.0104	0.0001	0.0014	0.0556	0.7060		
419	7	1.250	1.01	-0.0002	-0.0005	-0.1146	0.7933	-0.01	-1.05	0.06	-0.97	0.0894	0.6727		
	8	0.0033	-0.0000	0.0006	0.0005	-0.1146	0.7933	0.0162	0.0162	0.0002	0.0021	0.0556	0.6727		
	9	-0.0006	0.0002	-0.0001	-0.0001	-2.1399	1.0826	0.0149	0.0149	0.0001	0.0020	0.0556	0.6727		
419	7	1.250	3.16	-0.0003	-0.0004	-0.1383	0.7572	-0.01	-1.08	0.06	-0.98	0.0744	0.6763		
	8	0.0032	-0.0000	0.0006	0.0004	-0.1383	0.7572	0.0350	0.0350	0.0005	0.0047	0.0484	0.6712		
	9	-0.0007	0.0003	-0.0001	-0.0001	-2.2460	0.8827	0.0336	0.0336	0.0003	0.0045	0.0484	0.6712		
419	7	1.250	6.27	-0.0003	-0.0003	-0.2827	0.7195	-0.01	-1.13	0.06	-1.02	0.0644	0.6669		
	8	0.0024	-0.0001	0.0003	0.0003	-0.2827	0.7195	0.0616	0.0616	0.0007	0.0082	0.0363	0.6669		
	9	0.0000	0.0003	-0.0000	-0.0000	19.9981	-2.5969	0.0632	0.0632	0.0004	0.0084	0.0363	0.6669		

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key															
419	4	7 8 9	1.250 0.0020 -0.0005	2.43 -0.0001 0.0003	-0.00 0.0002 -0.0000	-0.2841 -2.8153	-0.01 0.7404 0.6848	1.17 0.0859 0.0887	0.06 0.0010 0.0006	1.04 0.0111 0.0114	0.0620 0.0384	0.6480 0.6440						
419	9	7 8 9	1.250 0.0016 -0.00014	12.64 -0.0001 0.0003	-0.00 0.0002 -0.0001	-0.3604 -1.1876	-0.02 0.7682 0.6782	1.20 0.1060 0.1092	0.06 0.0012 0.0008	1.07 0.0132 0.0135	0.0605 0.0401	0.6268 0.6212						
419	10	7 8 9	1.250 0.0029 -0.00007	-0.02 0.0000 0.0002	-0.00 0.0005 -0.0001	-0.1105 -1.8440	-0.01 0.8613 0.9414	1.03 0.0065 0.0062	0.06 0.0001 0.0000	0.96 0.0008 0.0008	0.0901 0.0108	0.6780 0.6911						
420	1	7 8 9	1.250 0.0020 -0.00013	-3.06 -0.0001 0.0002	-0.00 0.0003 -0.0002	-0.2907 -1.0309	-0.02 0.9060 0.7915	2.95 -0.0076 -0.0056	0.06 -0.0003 -0.0003	3.01 -0.0011 -0.0008	0.2029 0.3003	0.7324 0.7289						
420	2	7 8 9	1.250 0.0027 -0.00007	-1.02 0.0000 0.0002	-0.00 0.0004 -0.0001	-0.1617 -1.5791	-0.01 0.8117 1.1730	2.99 0.0085 0.0095	0.06 0.0001 0.0000	3.04 0.0012 0.0013	0.1131 0.0453	0.7151 0.6897						
420	3	7 8 9	1.250 0.0029 -0.00006	0.00 0.0000 0.0002	-0.00 0.0004 -0.0001	-0.1167 -2.1407	-0.01 0.8384 1.3367	3.01 0.0179 0.0181	0.06 0.0003 0.0002	3.06 0.0024 0.0024	0.0971 0.0615	0.6925 0.6842						
420	4	7 8 9	1.250 0.0030 -0.00005	0.54 -0.0000 0.0002	-0.00 0.0005 -0.0001	-0.1283 -2.6124	-0.01 0.8271 1.6629	3.01 0.0231 0.0228	0.06 0.0004 0.0002	3.06 0.0031 0.0031	0.0878 0.0611	0.6873 0.6801						
420	5	7 8 9	1.250 0.0032 -0.00007	1.08 -0.0000 0.0002	-0.00 0.0005 -0.0001	-0.1368 -2.0612	-0.02 0.8109 1.0490	3.02 0.0282 0.0276	0.06 0.0004 0.0003	3.07 0.0038 0.0037	0.0787 0.0546	0.6836 0.6736						
420	6	7 8 9	1.250 0.0033 -0.00006	3.15 -0.0000 0.0003	-0.00 0.0005 -0.0001	-0.1434 -2.3955	-0.01 0.7493 1.1650	3.05 0.0468 0.0463	0.06 0.0006 0.0003	3.08 0.0063 0.0062	0.0693 0.0430	0.6788 0.6751						
420	7	7 8 9	1.250 0.0024 0.0000	6.31 -0.0001 0.0003	-0.00 0.0003 -0.0000	-0.2479 38.0890	-0.01 0.8029 -7.9350	3.09 0.0721 0.0752	0.07 0.0009 0.0005	3.11 0.0095 0.0098	0.0642 0.0350	0.6636 0.6540						
420	8	7 8 9	1.251 0.0020 -0.00003	9.42 -0.0001 0.0003	-0.00 0.0003 -0.0001	-0.2549 -4.0933	-0.01 0.7414 1.6007	3.12 0.0950 0.0984	0.06 0.0011 0.0007	3.14 0.0121 0.0125	0.0601 0.0385	0.6411 0.6371						

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key														
420	9	1.240	12.66	-0.002	-0.000	-0.3064	-0.002	-0.001	-0.001	0.7996	0.1135	0.0013	0.0141	0.0599	0.6219	
	6	0.0017	-0.0001	0.0002	0.0002	-0.3668	0.7996	0.6851	0.1179	0.0009	0.1155	0.0009	0.0144	0.0389	0.6123	
	9	-0.0013	0.0003	-0.0001	-0.0001	-1.3668	0.6851	0.1179	0.0009	0.1155	0.0009	0.0144	0.0389	0.6123	0.6123	
420	10	1.249	-0.000	0.0005	-0.0005	-0.1159	-0.001	-0.001	3.01	0.0003	0.0177	0.0003	3.05	0.0879	0.6674	
	6	0.0029	0.0003	0.0001	0.0001	-1.6347	-0.8670	0.8670	0.0178	0.0001	0.0178	0.0001	0.0023	0.0541	0.6632	
	9	-0.0009	0.0003	-0.0001	-0.0001	-1.6347	0.8670	0.8670	0.0178	0.0001	0.0178	0.0001	0.0023	0.0541	0.6632	
421	1	1.252	-3.04	-0.003	-0.003	-0.2727	-0.012	0.9790	9.07	0.003	0.0245	0.0002	9.00	0.0786	0.7111	
	6	0.0022	0.0002	0.0002	0.0002	-1.0127	0.9790	0.0261	0.0261	0.0002	0.0261	0.0002	0.0036	0.0475	0.7065	
	9	-0.0012	0.0002	-0.0002	-0.0002	-1.0127	0.9790	0.0261	0.0261	0.0002	0.0261	0.0002	0.0036	0.0475	0.7065	
421	2	1.251	-0.96	0.004	0.004	-0.1526	-0.0175	1.1346	9.10	0.006	0.0447	0.0004	9.02	0.0688	0.6967	
	6	0.0026	0.0002	0.0002	0.0002	-1.3619	1.1346	1.1346	0.0455	0.0004	0.0455	0.0004	0.0062	0.0457	0.6882	
	9	-0.0009	0.0002	-0.0002	-0.0002	-1.3619	1.1346	1.1346	0.0455	0.0004	0.0455	0.0004	0.0062	0.0457	0.6882	
421	3	1.252	0.04	-0.005	-0.005	-0.1047	-0.0195	1.1377	9.11	0.006	0.0540	0.0004	9.04	0.0665	0.6856	
	6	0.0031	0.0002	0.0005	0.0005	-0.1047	0.0195	1.1377	0.0541	0.0004	0.0541	0.0004	0.0074	0.0443	0.6856	
	9	-0.0008	0.0002	-0.0001	-0.0001	-1.5619	1.1377	1.1377	0.0541	0.0004	0.0541	0.0004	0.0074	0.0443	0.6856	
421	4	1.252	0.57	0.005	0.005	-0.1057	-0.01	1.2078	9.12	0.006	0.0588	0.0004	9.05	0.0665	0.6831	
	6	0.0033	0.0002	0.0005	0.0005	-0.1057	0.01	1.2078	0.0588	0.0004	0.0588	0.0004	0.0080	0.0422	0.6825	
	9	-0.0007	0.0002	-0.0001	-0.0001	-1.7502	1.2078	1.2078	0.0588	0.0004	0.0588	0.0004	0.0080	0.0422	0.6825	
421	5	1.250	1.12	-0.002	-0.002	-0.1301	-0.01	1.2375	9.13	0.006	0.0632	0.0005	9.05	0.0665	0.6777	
	6	0.0033	0.0002	0.0005	0.0005	-0.1301	0.01	1.2375	0.0632	0.0005	0.0632	0.0005	0.0085	0.0413	0.6766	
	9	-0.0008	0.0002	-0.0002	-0.0002	-1.8025	1.2375	1.2375	0.0632	0.0005	0.0632	0.0005	0.0085	0.0413	0.6766	
421	6	1.251	3.22	0.004	0.004	-0.1514	-0.02	0.9760	9.15	0.006	0.0791	0.0006	9.07	0.0633	0.6605	
	6	0.0032	0.0003	0.0004	0.0004	-0.1514	0.02	0.9760	0.0791	0.0006	0.0791	0.0006	0.0104	0.0584	0.6554	
	9	-0.0008	0.0003	-0.0001	-0.0001	-1.9549	0.9760	0.9760	0.0801	0.0006	0.0801	0.0006	0.0105	0.0584	0.6554	
421	7	1.250	6.37	-0.003	-0.003	-0.2748	-0.02	5.4496	9.18	0.007	0.1004	0.0007	9.09	0.0587	0.6399	
	6	0.0024	0.0001	0.0003	0.0003	-0.2748	0.02	5.4496	0.1004	0.0007	0.1004	0.0007	0.0128	0.0370	0.6322	
	9	-0.0000	0.0003	-0.0000	-0.0000	-26.2595	5.4496	5.4496	0.1038	0.0007	0.1038	0.0007	0.0131	0.0370	0.6322	
421	8	1.251	9.48	0.003	0.003	-0.2729	-0.02	0.9621	9.21	0.007	0.1176	0.0010	9.11	0.0578	0.6202	
	6	0.0021	0.0001	0.0003	0.0003	-0.2729	0.02	0.9621	0.1176	0.0010	0.1176	0.0010	0.0145	0.0446	0.6100	
	9	-0.0006	0.0003	-0.0001	-0.0001	-2.3968	0.9621	0.9621	0.1204	0.0010	0.1204	0.0010	0.0146	0.0446	0.6100	
421	9	1.250	12.65	-0.002	-0.002	-0.2842	-0.01	0.8668	9.23	0.006	0.1300	0.0009	9.12	0.0534	0.6061	
	6	0.0017	0.0001	0.0002	0.0002	-0.2842	0.01	0.8668	0.1300	0.0009	0.1300	0.0009	0.0157	0.0358	0.5958	
	9	-0.0011	0.0003	-0.0001	-0.0001	-1.5262	0.8668	0.8668	0.1338	0.0009	0.1338	0.0009	0.0159	0.0358	0.5958	

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key															
421	10	7	1.250	0.0000	0.04	-0.0005	-0.0001	-0.0005	-0.1255	-0.8330	-0.02	9.11	0.0607	0.0007	9.04	0.0651	0.6831
		8	0.0030	-0.0003	0.0000	0.0005	0.0001	0.0005	0.1255	0.8330	0.02	0.0539	0.0004	0.0073	0.0073	0.0439	0.6812
		9	-0.0009	0.0003	-0.0003	-0.0001	-0.0001	-1.6407	-1.0175	1.0175	0.0539	0.0539	0.0004	0.0073	0.0073	0.0439	0.6812
422	3	7	1.249	-3.01	-0.0001	-0.0003	-0.0002	-0.3125	-0.9399	-0.03	12.04	0.05	0.0006	12.05	0.0745	0.7116	
		8	0.0019	-0.0001	0.0001	0.0003	0.0002	0.7190	0.9429	0.9399	0.0416	0.0006	0.0003	0.0059	0.0062	0.0442	0.7070
		9	-0.0011	0.0001	-0.0001	-0.0002	-0.0002	-0.7190	-0.9429	-0.9399	0.0442	0.0003	0.0003	0.0062	0.0062	0.0442	0.7070
422	4	7	1.251	-0.95	-0.0001	-0.0004	-0.0001	-0.1613	-0.8246	-0.03	12.07	0.06	0.0008	12.07	0.0718	0.6947	
		8	0.0027	-0.0000	0.0000	0.0004	0.0001	0.1613	0.8246	0.8246	0.0607	0.0008	0.0005	0.0084	0.0086	0.0421	0.6836
		9	-0.0007	0.0001	-0.0001	-0.0001	-1.2015	-1.1364	1.1364	1.1364	0.0629	0.0005	0.0005	0.0086	0.0086	0.0421	0.6836
422	5	7	1.250	0.10	-0.0001	-0.0004	-0.0001	-0.1315	-0.8100	-0.03	12.08	0.05	0.0009	12.08	0.0714	0.6801	
		8	0.0029	-0.0000	0.0000	0.0004	0.0001	0.1315	0.8100	0.8100	0.0696	0.0009	0.0005	0.0094	0.0096	0.0410	0.6766
		9	-0.0006	0.0001	-0.0001	-0.0001	-1.4851	-1.1751	1.1751	1.1751	0.0713	0.0005	0.0005	0.0096	0.0096	0.0410	0.6766
422	6	7	1.249	0.10	-0.0002	-0.0005	-0.0001	-0.1309	-0.8327	-0.03	12.08	0.06	0.0009	12.08	0.0713	0.6802	
		8	0.0031	-0.0000	0.0000	0.0005	0.0001	0.1309	0.8327	0.8327	0.0695	0.0009	0.0005	0.0094	0.0096	0.0408	0.6749
		9	-0.0006	0.0002	-0.0002	-0.0001	-1.6358	-1.2403	1.2403	1.2403	0.0712	0.0005	0.0005	0.0096	0.0096	0.0408	0.6749
422	7	7	1.248	1.11	-0.0002	-0.0005	-0.0001	-0.1294	-0.7427	-0.03	12.09	0.06	0.0010	12.09	0.0685	0.6688	
		8	0.0035	-0.0000	0.0000	0.0005	0.0001	0.1294	0.7427	0.7427	0.0778	0.0010	0.0006	0.0104	0.0105	0.0404	0.6636
		9	-0.0006	0.0002	-0.0002	-0.0001	-1.6962	-1.0638	1.0638	1.0638	0.0792	0.0006	0.0006	0.0105	0.0105	0.0404	0.6636
422	8	7	1.250	3.26	-0.0002	-0.0004	-0.0001	-0.1520	-0.7211	-0.04	12.11	0.06	0.0011	12.10	0.0626	0.6531	
		8	0.0032	-0.0000	0.0000	0.0004	0.0001	0.1520	0.7211	0.7211	0.0927	0.0011	0.0007	0.0121	0.0122	0.0376	0.6484
		9	-0.0006	0.0002	-0.0002	-0.0001	-1.9882	-0.8267	0.8267	0.8267	0.0948	0.0007	0.0007	0.0122	0.0122	0.0376	0.6484
422	9	7	1.250	6.37	-0.0001	-0.0003	-0.0000	-0.2908	-0.6967	-0.03	12.14	0.06	0.0013	12.12	0.0602	0.6305	
		8	0.0025	-0.0001	0.0001	0.0003	0.0000	0.2908	0.6967	0.6967	0.1114	0.0013	0.0008	0.0140	0.0143	0.0386	0.6221
		9	0.0002	0.0002	-0.0002	-0.0000	6.6334	0.6993	0.6993	0.6993	0.1155	0.0008	0.0008	0.0143	0.0143	0.0386	0.6221
422	10	7	1.249	9.49	-0.0001	-0.0003	-0.0000	-0.2448	-0.8164	-0.02	12.16	0.06	0.0014	12.14	0.0580	0.6136	
		8	0.0021	-0.0001	0.0001	0.0003	0.0000	0.2448	0.8164	0.8164	0.1253	0.0014	0.0010	0.0153	0.0152	0.0425	0.6003
		9	-0.0004	0.0002	-0.0002	-0.0000	-2.8645	0.3785	0.3785	0.3785	0.1272	0.0010	0.0010	0.0152	0.0152	0.0425	0.6003
422	11	7	1.249	12.66	-0.0002	-0.0003	-0.0001	-0.2531	-0.7716	-0.02	12.18	0.06	0.0015	12.15	0.0562	0.6007	
		8	0.0019	-0.0000	0.0000	0.0003	0.0001	0.2531	0.7716	0.7716	0.1348	0.0015	0.0010	0.0162	0.0165	0.0374	0.5926
		9	-0.0010	0.0002	-0.0002	-0.0001	-1.4202	0.7035	0.7035	0.7035	0.1395	0.0010	0.0010	0.0165	0.0165	0.0374	0.5926
422	12	7	1.249	0.06	-0.0002	-0.0005	-0.0001	-0.1306	-0.8440	-0.03	12.08	0.06	0.0009	12.08	0.0682	0.6757	
		8	0.0030	-0.0000	0.0000	0.0005	0.0001	0.1306	0.8440	0.8440	0.0692	0.0009	0.0005	0.0093	0.0095	0.0394	0.6717
		9	-0.0009	0.0002	-0.0002	-0.0001	-1.5199	0.7548	0.7548	0.7548	0.0707	0.0005	0.0005	0.0095	0.0095	0.0394	0.6717

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																			
423	1	7	1.051	-3.13	-0.00	-0.004	-0.1962	-0.03	-3.00	0.06	-3.01	0.1155	0.6797	0.1141	0.6817	0.0032	-0.0012	0.0002	0.0004	-0.0473	-0.0010	-0.0059
423	2	7	1.050	-1.08	0.0006	-0.0001	-0.1183	-0.03	-2.97	0.06	-2.98	0.1389	0.6771	0.1530	0.6910	0.0041	-0.0005	0.0002	0.0006	-0.0257	-0.0007	-0.0034
423	3	7	1.046	-0.04	0.0006	-0.0001	-0.0608	-0.03	-2.95	0.06	-2.98	0.1594	0.6886	0.1991	0.7037	0.0044	-0.0000	0.0002	0.0006	-0.0147	-0.0004	-0.0018
423	4	7	1.050	-0.03	0.0006	-0.0000	-0.0721	-0.03	-2.95	0.06	-2.97	0.1567	0.6832	0.2066	0.7224	0.0045	-0.0003	0.0002	0.0006	-0.0147	-0.0004	-0.0020
423	5	7	1.050	1.00	0.0006	-0.0000	-0.0578	-0.03	-2.94	0.06	-2.96	0.1898	0.7696	0.2988	0.7834	0.0051	-0.0001	0.0002	0.0006	-0.0045	-0.0001	-0.0007
423	6	7	1.047	3.10	0.0006	-0.0000	-0.0398	-0.03	-2.90	0.06	-2.94	0.2011	0.6431	0.1109	0.6401	0.0049	-0.0001	0.0002	0.0006	-0.0120	-0.0004	-0.0014
423	7	7	1.048	6.21	0.0004	-0.0000	-0.1387	-0.03	-2.86	0.06	-2.91	0.1313	0.6583	0.0920	0.6514	0.0034	-0.0007	0.0002	0.0004	-0.0416	-0.0010	-0.0057
423	8	7	1.051	9.31	0.0004	-0.0000	-0.1586	-0.02	-2.83	0.06	-2.88	0.0946	0.6559	0.0653	0.6548	0.0032	-0.0004	0.0002	0.0004	-0.0720	-0.0013	-0.0098
423	9	7	1.049	12.51	0.0001	-0.0000	-0.1688	-0.02	-2.80	0.05	-2.85	0.0812	0.6362	0.0546	0.6338	0.0030	-0.0009	0.0001	0.0004	-0.0990	-0.0016	-0.0130
423	10	7	1.047	-0.05	0.0006	-0.0000	-0.0648	-0.02	-2.95	0.06	-2.98	0.1598	0.6943	0.2046	0.7307	0.0045	-0.0004	0.0002	0.0006	-0.0149	-0.0004	-0.0020
424	1	7	1.049	-3.11	0.0004	-0.0001	-0.1623	-0.03	-1.00	0.06	-1.03	0.1339	0.6711	0.1411	0.6619	0.0033	-0.0001	0.0002	0.0004	-0.0342	-0.0009	-0.0045

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key →												
424	2	7	1.047	-1.07	-0.00	-0.005	-0.1145	0.7083	-0.03	-0.96	0.06	-1.02	0.1770	0.6688
		8	0.0042	-0.0000	0.0005	0.0000	-0.0000	0.7083	-0.03	-0.0123	-0.0004	-0.0016	0.2519	0.7240
		9	-0.0004	0.0002	-0.0001	-2.5107	1.2480	-0.0098	-0.0004	-0.0004	-0.0004	-0.0014		
424	3	7	1.048	-0.04	-0.00	-0.006	-0.0678	0.7250	-0.03	-0.95	0.06	-1.00	0.2702	0.7666
		8	0.0047	-0.0000	0.0006	0.0000	-0.0000	-0.0026	-0.0001	-0.0002	-0.0004	-0.0004	0.5306	0.9021
		9	0.0000	0.0002	-0.0001	15.0532	-8.2872	-0.0023	-0.0002	-0.0002	-0.0004	-0.0004		
424	4	7	1.047	0.50	-0.00	-0.00	-0.0608	-0.03	-0.94	0.06	-0.99	0.2417	0.6418	
		8	0.0051	-0.0000	0.0007	-0.0000	0.6869	0.0015	0.0000	0.0000	0.0002	-0.3302	0.3833	
		9	-0.0000	0.0002	-0.0000	-21.0146	6.8162	0.0017	-0.0001	-0.0001	0.0001			
424	5	7	1.050	1.00	-0.00	-0.00	-0.0541	-0.03	-0.93	0.06	-0.99	0.2323	0.6468	
		8	0.0052	-0.0000	0.0007	0.0000	0.6863	0.0052	0.0002	0.0000	0.0006	0.0151	0.5859	
		9	-0.0002	0.0002	-0.0000	-4.4329	1.1797	0.0049	0.0000	0.0000	0.0005			
424	6	7	1.044	3.10	-0.00	-0.00	-0.0474	-0.02	-0.91	0.06	-0.97	0.1494	0.6506	
		8	0.0050	-0.0000	0.0006	0.0000	0.6215	0.0256	0.0007	0.0004	0.0033	0.0990	0.6506	
		9	-0.0002	0.0002	-0.0000	-5.4027	1.4673	0.0236	0.0004	0.0004	0.0030			
424	7	7	1.048	6.21	-0.00	-0.00	-0.1354	-0.02	-0.87	0.06	-0.93	0.1112	0.6702	
		8	0.0033	-0.0000	0.0004	0.0000	0.6754	0.0551	0.0012	0.0008	0.0073	0.0746	0.6574	
		9	0.0010	0.0002	-0.0000	1.0972	0.3314	0.0574	0.0008	0.0008	0.0075			
424	8	7	1.049	9.34	-0.00	-0.00	-0.1548	-0.02	-0.83	0.06	-0.91	0.0880	0.6486	
		8	0.0031	-0.0000	0.0004	0.0000	0.7394	0.0839	0.0014	0.0009	0.0108	0.0564	0.6460	
		9	0.0002	0.0002	-0.0000	4.6074	1.0057	0.0877	0.0009	0.0009	0.0113			
424	9	7	1.052	12.54	-0.00	-0.00	-0.1630	-0.02	-0.80	0.05	-0.88	0.0833	0.6279	
		8	0.0031	-0.0001	0.0004	0.0000	0.6741	0.1074	0.0017	0.0013	0.0134	0.0592	0.6172	
		9	-0.0008	0.0002	-0.0001	-1.3325	0.7421	0.1107	0.0013	0.0013	0.0136			
424	10	7	1.047	-0.04	-0.00	-0.00	-0.0598	-0.02	-0.95	0.06	-1.00	0.3032	0.8488	
		8	0.0045	-0.0000	0.0007	0.0000	0.7149	-0.0026	-0.0001	-0.0002	-0.0004	0.5281	0.9952	
		9	-0.0002	0.0002	-0.0000	-5.2530	1.3802	-0.0024	-0.0002	-0.0002	-0.0004			
425	1	7	1.047	-3.11	-0.00	-0.00	-0.1474	-0.02	-0.10	0.05	-0.06	0.1449	0.6608	
		8	0.0033	-0.0000	0.0005	0.0000	0.7557	-0.0281	-0.0008	-0.0007	-0.0037	0.1634	0.6816	
		9	-0.0010	0.0002	-0.0001	-1.0989	0.6645	-0.0236	-0.0007	-0.0007	-0.0032			
425	2	7	1.049	-1.09	-0.00	-0.00	-0.1101	-0.02	-0.07	0.06	-0.03	0.2127	0.6716	
		8	0.0042	-0.0000	0.0006	0.0000	0.7087	-0.0070	-0.0003	-0.0003	-0.0007	0.3248	0.7364	
		9	-0.0005	0.0002	-0.0000	-2.2895	0.7458	-0.0052	-0.0003	-0.0003	-0.0007			

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key															
425	3	7 8 9	1.048 0.0046 -0.00001	-0.0002 0.0000 0.00002	-0.0006 0.0000 -0.0000	-0.0721 -7.9189	-0.02 0.7273 2.9309	-0.04 0.0019 0.0029	0.06 0.0000 -0.0000	-0.002 0.0003 0.0003	0.1041 -0.1182	0.5086 0.5675						
425	4	7 8 9	1.048 0.0049 -0.00000	0.0000 0.0000 0.00002	0.0006 0.0000 -0.0000	-0.0646 -11.9560	-0.02 0.6881 4.2726	-0.04 0.0058 0.0063	0.06 0.0002 0.0000	-0.001 0.0008 0.0008	0.2212 0.0614	0.6094 0.6548						
425	5	7 8 9	1.048 0.0050 -0.00001	0.0000 0.0000 0.00002	0.0006 0.0000 -0.0000	-0.0587 -8.9425	-0.03 0.6651 2.8451	-0.02 0.0100 0.0101	0.06 0.0003 0.0001	-0.0013 0.0013 0.0013	0.1972 0.0829	0.6094 0.6547						
425	6	7 8 9	1.048 0.0050 -0.00002	0.0000 0.0000 0.00002	0.0006 0.0000 -0.0000	-0.0408 -5.2041	-0.02 0.6232 0.6100	0.00 0.0312 0.0299	0.06 0.0008 0.0005	0.01 0.0041 0.0038	0.1387 0.0910	0.6597 0.6515						
425	7	7 8 9	1.046 0.0034 0.00007	0.0000 0.0000 0.00002	0.0004 0.0000 0.00001	-0.1210 1.7321	-0.02 0.6981 0.8023	0.04 0.0611 0.0638	0.06 0.0012 0.0009	0.04 0.0081 0.0084	0.1035 0.0710	0.6669 0.6633						
425	8	7 8 9	1.048 0.0032 -0.00000	0.0000 0.0000 0.00002	0.0004 0.0000 -0.0000	-0.1619 -38.7555	-0.02 0.6691 -12.1304	0.07 0.0885 0.0925	0.06 0.0015 0.0010	0.06 0.0114 0.0118	0.0865 0.0568	0.6455 0.6399						
425	9	7 8 9	1.049 0.0031 -0.00009	0.0000 0.0000 0.00002	0.0004 0.0000 -0.00001	-0.1637 -1.1171	-0.02 0.6483 0.6705	0.11 0.1106 0.1130	0.05 0.0019 0.0015	0.09 0.0137 0.0138	0.0862 0.0664	0.6225 0.6116						
425	10	7 8 9	1.049 0.0046 -0.00004	0.0000 0.0000 0.00002	0.0006 0.0000 -0.0000	-0.1721 -3.1816	-0.02 0.7076 0.5861	-0.04 0.0018 0.0028	0.06 0.0000 0.0000	-0.002 0.0002 0.0002	0.0963 -0.1394	0.5724 0.5153						
426	1	7 8 9	1.047 0.0033 -0.00010	0.0000 0.0000 0.00002	0.0004 0.0000 -0.00001	-0.1563 -1.1048	-0.02 0.7390 0.6143	0.97 -0.0203 -0.0171	0.05 -0.0006 -0.0006	0.93 -0.0026 -0.0023	0.1700 0.1979	0.6580 0.6844						
426	2	7 8 9	1.048 0.0044 -0.00002	0.0000 0.0000 0.00002	0.0006 0.0000 -0.0000	-0.1072 -3.7314	-0.03 0.7041 1.7259	1.01 -0.0005 -0.0004	0.06 -0.0000 -0.0001	0.95 -0.0000 0.0000	0.7169 -1.8296	0.4787 0.0587						
426	3	7 8 9	1.049 0.0046 -0.00001	0.0000 0.0000 0.00002	0.0006 0.0000 -0.0000	-0.0643 -8.0795	-0.02 0.7365 3.1841	1.02 0.0070 0.0076	0.06 0.0002 0.0000	0.97 0.0009 0.0010	0.1957 0.0576	0.7025 0.6609						

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																										
426	4	7	1.048	0.52	-0.000	0.0006	-0.0511	0.7015	-0.02	1.03	0.06	0.97	0.1750	0.6900	8	0.0048	-0.0002	0.0000	0.0006	0.0000	0.0000	0.0000	0.0004	0.0004	0.0016	0.0895	0.6684	
426	5	7	1.049	1.03	-0.000	0.0006	-0.0555	0.6482	-0.02	1.04	0.06	0.98	0.1630	0.6767	8	0.0052	-0.0002	0.0000	0.0006	0.0000	0.0000	0.0000	0.0004	0.0004	0.0022	0.0877	0.6591	
426	6	7	1.047	3.13	-0.000	0.0006	-0.0438	0.6303	-0.02	1.07	0.06	1.01	0.1276	0.6710	8	0.0049	0.0002	0.0000	0.0006	0.0000	0.0000	0.0000	0.0004	0.0004	0.0049	0.0850	0.6641	
426	7	7	1.048	6.24	-0.000	0.0004	-0.1233	0.7008	-0.02	1.11	0.06	1.03	0.0977	0.6671	8	0.0033	0.0002	0.0001	0.0004	0.0000	0.0000	0.0000	0.0004	0.0004	0.0093	0.0652	0.6580	
426	8	7	1.050	9.35	-0.000	0.0004	-0.1619	0.6691	-0.02	1.14	0.06	1.06	0.0862	0.6415	8	0.0032	0.0002	0.0000	0.0004	0.0000	0.0000	0.0000	0.0004	0.0004	0.0120	0.0569	0.6338	
426	11	7	1.044	12.53	-0.000	0.0004	-0.1552	0.7265	-0.02	1.16	0.05	1.07	0.0899	0.6121	8	0.0031	0.0000	0.0001	0.0004	0.0000	0.0000	0.0000	0.0004	0.0004	0.0140	0.0668	0.6040	
426	12	7	1.055	0.03	-0.000	0.0000	-0.0709	0.7625	-0.03	1.01	0.05	0.96	0.1731	0.6231	8	0.0045	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0009	0.0332	0.5922	
427	1	7	1.050	3.08	-0.000	0.0005	-0.1606	0.7997	-0.02	2.96	0.04	3.03	0.3683	0.6731	8	0.0033	0.0001	0.0000	0.0005	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.3943	0.7231
427	2	7	1.048	1.05	-0.000	0.0006	-0.1133	0.7467	-0.02	2.99	0.05	3.05	0.1760	0.7043	8	0.0042	0.0000	0.0000	0.0006	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0658	0.6660	
427	3	7	1.053	0.00	-0.000	0.0006	-0.0534	0.7486	-0.02	3.00	0.05	3.06	0.1421	0.6594	8	0.0046	0.0000	0.0000	0.0006	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0658	0.6660	
427	4	7	1.048	0.52	-0.000	0.0006	-0.0510	0.6996	-0.02	3.01	0.05	3.07	0.1336	0.6676	8	0.0049	0.0000	0.0000	0.0006	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0658	0.6660	

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key														
427	5	7 8 9	1.048 0.0050 -0.0002	-0.0000 1.03 0.0002	-0.0006 -0.0000 -0.0000	-0.0447 -4.0392	-0.925 1.5387	3.02 0.0307 0.0301	0.05 0.0007 0.0004	0.0040 0.0039	0.1263 0.0806	0.6646 0.6602					
427	6	7 8 9	1.048 0.0046 0.0000	-0.0000 5.15 0.0000	-0.0006 -0.0000 -0.0000	-0.0372 23.7054	-0.02 0.7115 -7.0987	3.05 0.0523 0.0516	0.04 0.0011 0.0007	0.0077 0.0069	0.1052 0.0716	0.6752 0.6713					
427	7	7 8 9	1.051 0.0033 0.0011	6.25 -0.0000 0.0002	-0.0004 0.0000 0.0000	-0.1339 0.9229	-0.02 0.2339 0.3897	3.08 0.0795 0.0835	0.05 0.0014 0.0009	0.0104 0.0108	0.0904 0.0588	0.6547 0.6506					
427	8	7 8 9	1.049 0.0031 0.0000	9.35 -0.0001 0.0002	-0.0004 0.0000 0.0000	-0.1582 11.7881	-0.02 0.7097 13.4931	3.12 0.1025 0.1058	0.05 0.0018 0.0014	0.0129 0.0130	0.0887 0.0662	0.6292 0.6177					
427	9	7 8 9	1.045 0.0030 -0.0007	12.56 -0.0000 0.0002	-0.0004 0.0000 -0.0001	-0.1527 -1.4108	-0.01 0.7176 0.8426	3.14 0.1198 0.1217	0.04 0.0019 0.0015	0.0145 0.0144	0.0809 0.0630	0.6052 0.5933					
427	10	7 8 9	1.048 0.0045 -0.0004	-0.0000 0.0000 0.0002	-0.0006 0.0000 -0.0000	-0.0570 -3.0225	-0.01 0.7371	3.01 0.0194 0.0204	0.05 0.0005 0.0003	0.0025 0.0026	0.1335 0.0755	0.6529 0.6444					
428	1	7 8 9	1.053 0.0033 -0.0006	-3.06 -0.0000 0.0001	-0.0005 0.0000 -0.0001	-0.1457 -1.5288	-0.02 0.8170 1.1328	6.01 0.0088 0.0111	0.04 0.0001 0.0000	0.0011 0.0014	0.1086 0.0439	0.6683 0.6630					
428	2	7 8 9	1.050 0.0042 -0.0002	-1.02 -0.0000 0.0001	-0.0006 0.0000 -0.0001	-0.1023 -3.8912	-0.01 0.7312 2.4302	6.05 0.0289 0.0308	0.05 0.0006 0.0004	0.0039 0.0041	0.1191 0.0723	0.6816 0.6695					
428	3	7 8 9	1.050 0.0044 -0.0001	0.00 -0.0000 0.0001	-0.0006 0.0000 -0.0000	-0.0383 -5.6703	-0.01 0.7768 2.8074	6.06 0.0395 0.0403	0.04 0.0008 0.0005	0.0054 0.0054	0.1124 0.0734	0.6835 0.6754					
428	4	7 8 9	1.045 0.0050 -0.0000	-0.0000 0.53 -0.0001	-0.0006 0.0000 -0.0000	-0.0414 -35.7877	-0.01 0.6946 16.0542	6.06 0.0459 0.0463	0.05 0.0009 0.0006	0.0062 0.0062	0.1045 0.0684	0.6790 0.6712					
428	5	7 8 9	1.049 0.0051 -0.0000	1.07 -0.0000 0.0002	-0.0007 0.0000 -0.0000	-0.0450 -17.4742	-0.01 0.6830 6.1064	6.08 0.0509 0.0507	0.04 0.0010 0.0006	0.0069 0.0068	0.0999 0.0657	0.6804 0.6768					

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key													
428	6	7	1.046	5.19	-0.000	-0.006	-0.0374	-0.01	6.10	0.05	6.07	0.0905	0.6666		
		6	0.0045	-0.0000	0.0006	0.0374	0.7049	0.0713	0.0012	0.0095	0.0585	0.6657			
		9	-0.0000	0.0002	-0.0000	-11.8169	0.5004	0.0717	0.0008	0.0095	0.0585	0.6657			
428	7	7	1.047	6.26	-0.000	-0.004	-0.1192	-0.01	6.13	0.05	6.10	0.0881	0.6400		
		8	0.0033	-0.0000	0.0004	0.1192	0.7286	0.0947	0.0016	0.0121	0.0594	0.6327			
		9	0.0009	0.0002	0.0000	1.2380	0.5022	0.0993	0.0011	0.0125	0.0594	0.6327			
428	8	7	1.049	9.39	-0.000	-0.004	-0.1389	-0.01	6.16	0.05	6.11	0.0908	0.6109		
		8	0.0031	-0.0000	0.0004	0.1389	0.7375	0.1111	0.0020	0.0135	0.0691	0.6020			
		9	0.0000	0.0002	0.0000	30.0622	2.3389	0.1144	0.0015	0.0137	0.0691	0.6020			
428	9	7	1.048	12.56	-0.000	-0.004	-0.1308	-0.01	6.17	0.04	6.12	0.0820	0.5962		
		8	0.0032	-0.0000	0.0004	0.1308	0.7279	0.1259	0.0020	0.0150	0.0605	0.5883			
		9	-0.0005	0.0002	-0.0001	-1.8303	0.9076	0.1299	0.0015	0.0152	0.0605	0.5883			
428	10	7	1.050	0.03	-0.000	-0.006	-0.01	6.06	0.04	6.04	0.1091	0.6772			
		8	0.0044	-0.0000	0.0006	0.0497	0.7641	0.0399	0.0008	0.0054	0.0717	0.6705			
		9	-0.0003	0.0002	-0.0000	-3.1800	1.0000	0.0405	0.0005	0.0054	0.0717	0.6705			
429	1	7	1.045	-3.01	-0.000	-0.000	-0.01	9.08	0.04	9.01	0.1089	0.6965			
		8	0.0032	-0.0001	0.0005	-0.1523	0.7926	0.0280	0.0006	0.0039	0.0631	0.6799			
		9	-0.0005	0.0001	-0.0001	-1.7198	1.6025	0.0305	0.0003	0.0041	0.0631	0.6799			
429	2	7	1.049	-0.99	-0.000	-0.006	-0.01	9.11	0.04	9.03	0.0945	0.6849			
		8	0.0042	-0.0000	0.0006	0.0895	0.7319	0.0495	0.0009	0.0067	0.0598	0.6782			
		9	-0.0002	0.0001	-0.0001	-3.5901	1.8925	0.0511	0.0006	0.0069	0.0598	0.6782			
429	3	7	1.046	0.04	-0.000	-0.007	-0.01	9.13	0.04	9.04	0.0889	0.6808			
		8	0.0046	-0.0000	0.0007	-0.0498	0.7487	0.0603	0.0010	0.0082	0.0568	0.6753			
		9	-0.0001	0.0001	-0.0000	-5.1440	1.8703	0.0614	0.0006	0.0082	0.0568	0.6753			
429	4	7	1.046	0.55	-0.000	-0.007	-0.01	9.14	0.04	9.04	0.0879	0.6769			
		8	0.0052	-0.0000	0.0007	-0.0478	0.6887	0.0556	0.0011	0.0089	0.0556	0.6751			
		9	-0.0000	0.0001	-0.0000	-24.9102	7.2526	0.0663	0.0007	0.0089	0.0556	0.6751			
429	5	7	1.043	1.10	-0.000	-0.007	-0.01	9.14	0.04	9.05	0.0859	0.6710			
		8	0.0052	-0.0000	0.0007	-0.0455	0.6662	0.0706	0.0012	0.0094	0.0551	0.6662			
		9	-0.0000	0.0001	-0.0000	-31.8203	8.3177	0.0711	0.0007	0.0094	0.0551	0.6662			
429	6	7	1.045	3.20	-0.000	-0.006	-0.01	9.17	0.05	9.06	0.0850	0.6495			
		8	0.0044	-0.0002	0.0006	-0.0245	0.7474	0.0883	0.0015	0.0114	0.0545	0.6495			
		9	-0.0000	0.0002	0.0000	-59.0703	-5.0815	0.0902	0.0009	0.0116	0.0545	0.6495			

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																												
429	7	1.046	6.28	-0.00	-0.00	-0.01	9.20	0.05	9.09	0.0976	0.6204	0.0033	-0.0000	0.0004	0.0001	-0.1192	0.7286	0.1053	0.0020	0.0130	0.0754	0.6066	0.0007	0.0003	0.0001	2.1297	0.8775	0.1082	0.0016	0.0131
429	8	1.043	9.40	-0.00	-0.00	-0.01	9.21	0.04	9.10	0.0866	0.6013	0.0030	-0.0000	0.0004	0.0000	-0.1353	0.7759	0.1203	0.0020	0.0144	0.0629	0.5886	-0.0000	0.0002	0.0000	-13.4812	-1.0673	0.1239	0.0015	0.0145
429	9	1.047	12.56	-0.00	-0.00	-0.01	9.22	0.04	9.11	0.0711	0.5933	0.0030	-0.0000	0.0004	0.0000	-0.1531	0.7044	0.1341	0.0019	0.0159	0.0499	0.5824	-0.0008	0.0002	-0.0000	-1.2015	0.3706	0.1383	0.0013	0.0161
429	10	1.048	0.03	-0.00	-0.00	-0.01	9.13	0.04	9.04	0.0892	0.6803	0.0046	-0.0000	0.0007	-0.0351	0.7695	0.0598	0.0010	0.0081	0.0681	0.0551	0.6703	-0.0003	0.0002	-0.0000	-2.9439	0.7381	0.0610	0.0006	0.0081
430	1	1.049	-3.01	-0.00	-0.00	-0.01	12.04	0.04	12.06	0.0906	0.6955	0.0032	-0.0001	0.0005	-0.1584	0.7929	0.0481	0.0008	0.0066	0.0906	0.6955	0.6865	-0.0007	0.0001	-0.0001	-1.2471	0.8864	0.0513	0.0005	0.0070
430	2	1.050	-0.93	-0.00	-0.00	-0.01	12.07	0.04	12.08	0.0816	0.6786	0.0043	-0.0000	0.0006	-0.0926	0.7288	0.0682	0.0011	0.0092	0.0548	0.6786	0.6711	-0.0002	0.0001	-0.0000	-4.9494	1.9441	0.0710	0.0007	0.0095
430	3	1.048	0.06	-0.00	-0.00	-0.01	12.08	0.04	12.08	0.0822	0.6660	0.0050	-0.0000	0.0007	-0.0510	0.6996	0.0781	0.0012	0.0104	0.0522	0.6660	0.6591	-0.0001	0.0002	-0.0000	-5.5980	1.5309	0.0803	0.0008	0.0105
430	4	1.048	0.59	-0.00	-0.00	-0.01	12.08	0.04	12.09	0.0830	0.6612	0.0052	-0.0000	0.0006	-0.0519	0.6651	0.0829	0.0013	0.0109	0.0518	0.6612	0.6559	-0.0001	0.0002	-0.0000	-8.6371	1.8205	0.0850	0.0008	0.0111
430	5	1.049	1.10	-0.00	-0.00	-0.01	12.09	0.04	12.09	0.0835	0.6560	0.0051	-0.0000	0.0006	-0.0458	0.6596	0.0868	0.0014	0.0113	0.0520	0.6560	0.6518	-0.0000	0.0001	-0.0000	-4.2716	1.0.2369	0.0890	0.0009	0.0116
430	6	1.048	1.62	-0.00	-0.00	-0.01	12.10	0.04	12.10	0.0836	0.6493	0.0051	-0.0000	0.0006	-0.0420	0.6771	0.0912	0.0015	0.0118	0.0505	0.6493	0.6444	-0.0000	0.0002	-0.0000	-11.3831	0.9070	0.0938	0.0009	0.0120
430	7	1.047	2.15	-0.00	-0.00	-0.01	12.10	0.04	12.10	0.0847	0.6440	0.0049	-0.0000	0.0006	-0.0386	0.6805	0.0948	0.0016	0.0122	0.0523	0.6440	0.6431	-0.0000	0.0002	-0.0000	-11.8169	0.5004	0.0976	0.0010	0.0125

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key											
430	8	7	1.048	2.66	-0.006	-0.0422	-0.01	12.11	0.04	12.11	0.0017	0.0876	0.6377	
		8	0.0049	-0.0000	0.0000	-0.0422	0.6718	0.0979	0.0017	0.0124	0.0128	0.0560	0.6373	
		9	-0.0001	0.0002	0.0000	-9.7750	-0.6395	0.1007	0.0011	0.0128	0.0128	0.0560	0.6373	
430	9	7	1.049	5.20	-0.006	-0.0377	-0.01	12.11	0.05	12.11	0.0018	0.0901	0.6312	
		8	0.0045	-0.0000	0.0000	-0.0377	0.6960	0.1009	0.0018	0.0127	0.0128	0.0636	0.6312	
		9	0.0001	0.0002	0.0000	-9.5797	1.0931	0.1030	0.0013	0.0128	0.0128	0.0636	0.6312	
430	10	7	1.044	5.69	-0.005	-0.0573	-0.01	12.11	0.05	12.12	0.0019	0.0934	0.6271	
		8	0.0042	-0.0000	0.0000	-0.0573	0.6768	0.1038	0.0019	0.0130	0.0129	0.0720	0.6271	
		9	-0.0002	0.0002	0.0000	-4.5324	-0.9380	0.1052	0.0015	0.0129	0.0129	0.0720	0.6271	
430	11	7	1.047	4.23	-0.005	-0.0815	-0.01	12.12	0.05	12.13	0.0020	0.0954	0.6194	
		8	0.0042	-0.0000	0.0000	-0.0815	0.6617	0.1053	0.0020	0.0130	0.0130	0.0746	0.6194	
		9	0.0001	0.0002	0.0000	7.7357	1.8757	0.1074	0.0016	0.0131	0.0131	0.0746	0.6194	
430	12	7	1.044	6.30	-0.004	-0.1077	-0.01	12.13	0.05	12.13	0.0019	0.0868	0.6075	
		8	0.0033	-0.0000	0.0000	-0.1077	0.7413	0.1147	0.0019	0.0139	0.0139	0.0671	0.6075	
		9	0.0002	0.0003	0.0001	7.7567	3.0138	0.1185	0.0015	0.0141	0.0141	0.0671	0.6075	
430	13	7	1.047	9.39	-0.004	-0.1093	-0.01	12.15	0.05	12.14	0.0015	0.0795	0.5949	
		8	0.0032	-0.0000	0.0000	-0.1093	0.7579	0.1281	0.0020	0.0152	0.0154	0.0572	0.5949	
		9	-0.0002	0.0002	0.0000	-5.1841	-0.8242	0.1321	0.0015	0.0154	0.0154	0.0572	0.5949	
430	14	7	1.047	12.59	-0.004	-0.1308	-0.01	12.15	0.04	12.14	0.0012	0.0623	0.5919	
		8	0.0032	-0.0000	0.0000	-0.1308	0.7279	0.1402	0.0017	0.0166	0.0166	0.0425	0.5919	
		9	-0.0006	0.0002	-0.0000	-1.8306	0.4248	0.1435	0.0012	0.0166	0.0166	0.0425	0.5919	
430	15	7	1.045	0.07	-0.006	-0.0541	-0.01	12.07	0.05	12.08	0.0008	0.0803	0.6639	
		8	0.0046	-0.0000	0.0000	-0.0541	0.7312	0.0783	0.0012	0.0104	0.0105	0.0521	0.6639	
		9	-0.0004	0.0002	-0.0000	-2.6270	0.4150	0.0805	0.0008	0.0105	0.0105	0.0521	0.6639	
431	2	7	1.047	-2.98	-0.005	-0.1681	-0.02	15.09	0.05	15.07	0.0007	0.0792	0.6812	
		8	0.0031	-0.0001	0.0000	-0.1681	0.6121	0.0679	0.0010	0.0092	0.0096	0.0502	0.6812	
		9	-0.0010	0.0001	-0.0001	-0.6736	0.7989	0.0708	0.0007	0.0096	0.0096	0.0502	0.6812	
431	3	7	1.052	-0.96	-0.006	-0.0851	-0.02	15.11	0.05	15.09	0.0008	0.0798	0.6598	
		8	0.0043	-0.0000	0.0000	-0.0851	0.7110	0.0860	0.0013	0.0113	0.0113	0.0505	0.6598	
		9	-0.0004	0.0001	-0.0000	-2.0878	1.1144	0.0889	0.0008	0.0117	0.0117	0.0505	0.6598	
431	4	7	1.048	0.09	-0.006	-0.0550	-0.01	15.13	0.06	15.10	0.0009	0.0831	0.6492	
		8	0.0049	-0.0000	0.0000	-0.0550	0.6830	0.0940	0.0015	0.0122	0.0122	0.0500	0.6492	
		9	-0.0001	0.0001	-0.0000	-5.0834	2.1424	0.0973	0.0009	0.0126	0.0126	0.0500	0.6492	

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																					
431	5	7	1.048	0.0051	-0.0002	0.69	-0.0000	0.0001	-0.00	-0.006	-0.0421	0.6737	1.3897	0.0973	15.13	0.0017	0.05	15.10	0.0124	0.0129	0.0882	0.6397	0.6417
431	6	7	1.047	0.0051	-0.0003	1.14	-0.0000	0.0001	-0.00	-0.006	-0.0427	0.6602	0.9300	0.1003	15.14	0.0018	0.05	15.11	0.0126	0.0130	0.0913	0.6302	0.6305
431	7	7	1.048	0.0045	-0.0002	3.23	-0.0000	0.0002	-0.00	0.0006	-0.0374	0.7049	-0.1213	0.1104	15.15	0.0020	0.06	15.12	0.0134	0.0137	0.0931	0.6111	0.6088
431	8	7	1.048	0.0033	-0.0002	6.30	-0.0000	0.0002	-0.00	0.0004	-0.1132	0.7411	0.6581	0.1240	15.16	0.0020	0.06	15.13	0.0148	0.0151	0.0822	0.5979	0.5941
431	9	7	1.045	0.0033	-0.0003	9.39	-0.0000	0.0002	-0.00	0.0004	-0.1293	0.6882	0.3731	0.1353	15.17	0.0018	0.06	15.13	0.0160	0.0161	0.0680	0.5914	0.5850
431	10	7	1.047	0.0031	-0.0007	12.57	-0.0000	0.0002	-0.00	0.0004	-0.1496	0.7013	0.6638	0.1419	15.17	0.0016	0.05	15.14	0.0166	0.0168	0.0571	0.5881	0.5804
431	11	7	1.056	0.0047	-0.0005	0.69	-0.0000	0.0002	-0.00	0.0006	-0.0548	0.7031	0.4894	0.0965	15.12	0.0009	0.05	15.10	0.0120	0.0125	0.0805	0.6457	0.6472
432	1	7	0.899	0.0042	-0.0015	3.12	-0.0000	0.0001	-0.00	0.0006	-0.1123	0.7408	0.6634	-0.0479	-3.01	0.0015	0.05	-2.99	-0.0065	-0.0064	0.1503	0.6537	0.6701
432	2	7	0.900	0.0047	-0.0009	1.07	-0.0000	0.0001	-0.00	0.0006	-0.0035	0.7356	0.5028	-0.0284	-2.98	0.0007	0.05	-2.97	-0.0039	-0.0036	0.1333	0.6906	0.6939
432	3	7	0.898	0.0049	-0.0006	0.05	-0.0000	0.0001	-0.00	0.0007	0.0441	0.7160	0.5114	-0.0158	-2.97	0.0005	0.05	-2.96	-0.0024	-0.0022	0.1517	0.7266	0.7170
432	4	7	0.898	0.0050	-0.0004	0.51	-0.0000	0.0002	-0.00	0.0007	0.0794	0.7060	0.6613	-0.0105	-2.96	0.0004	0.05	-2.96	-0.0016	-0.0015	0.1907	0.7653	0.7549

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd																			
432	5		0.2100	0.0001	0.0002	0.0007	0.0002	0.0002	0.0007	0.0007	0.0007	-0.0007	0.1031	-0.0051	-0.0002	0.05	-2.93	-0.0003	-0.0009	0.2752	0.8485
432	6		0.9000	0.0002	0.0002	0.0009	0.0002	0.0002	0.0006	0.0001	0.0006	0.1165	-0.0135	0.0003	0.05	-2.93	0.0018	0.0017	0.1236	0.6912	
432	7		0.8999	0.0002	0.0003	0.0002	0.0003	0.0002	0.0004	0.0000	0.0004	-0.0697	-0.0455	0.0010	0.06	-2.89	0.0060	0.0062	0.1375	0.6654	
432	8		0.8999	0.0002	0.0002	0.0003	0.0002	0.0003	0.0004	0.0000	0.0004	-0.1374	-0.0758	0.0016	0.05	-2.87	0.0094	0.0097	0.1325	0.6211	
432	9		0.9000	0.0001	0.0001	0.0001	0.0001	0.0001	0.0004	0.0001	0.0004	-0.1542	-0.1006	0.0019	0.05	-2.84	0.0121	0.0122	0.1167	0.6017	
432	10		0.8999	0.0007	0.0002	0.0007	0.0002	0.0002	0.0006	0.0000	0.0006	0.0297	-0.0154	-0.0005	0.05	-2.97	-0.0024	-0.0022	0.1488	0.7290	
433	1		0.9000	0.0001	0.0001	0.0001	0.0001	0.0001	0.0005	0.0001	0.0005	-0.1427	-0.0477	0.0014	0.05	-1.04	-0.0062	-0.0060	0.1472	0.6584	
433	2		0.8999	0.0001	0.0001	0.0001	0.0001	0.0001	0.0006	0.0001	0.0006	-0.1002	-0.0370	-0.0010	0.05	-1.03	-0.0050	-0.0046	0.1432	0.6804	
433	3		0.8999	0.0007	0.0001	0.0007	0.0001	0.0001	0.0006	0.0001	0.0006	0.0089	-0.0133	-0.0004	0.05	-1.01	-0.0019	-0.0017	0.1726	0.7124	
433	4		0.8977	0.0007	0.0001	0.0007	0.0001	0.0001	0.0006	0.0001	0.0006	0.0034	-0.0134	-0.0004	0.05	-1.00	-0.0019	-0.0017	0.1734	0.7143	
433	5		0.8998	0.0006	0.0001	0.0006	0.0001	0.0001	0.0006	0.0001	0.0006	0.0516	-0.0024	-0.0002	0.05	-1.00	-0.0004	-0.0003	0.4427	0.9439	

See Key

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key →																			
433	6	7 8 9	0.900 0.0049 -0.0005	-0.005 0.0000 0.0001	-0.00 0.0007 -0.0001	0.0614 -1.7512	-0.02 0.7130 0.9996	-0.96 -0.0023 -0.0022	0.05 -0.0002 -0.0002	-1.00 0.0004 -0.0003	0.4688 0.6132	0.9733 0.6712										
433	7	7 8 9	0.898 0.0051 -0.0005	-0.003 0.0000 0.0001	-0.00 0.0006 -0.0000	0.0425 -1.6980	-0.02 0.6495 0.7527	-0.96 -0.0024 -0.0022	0.05 -0.0002 -0.0002	-1.00 0.0004 -0.0003	0.4428 0.6162	0.9232 0.6574										
433	8	7 8 9	0.899 0.0050 -0.0005	0.55 0.0000 0.0001	-0.00 0.0007 -0.0001	0.0819 -1.8131	-0.01 0.6988 1.0155	-0.96 0.0019 0.0011	0.05 -0.0000 -0.0000	-0.99 0.0002 0.0001	-0.0376 -0.3812	0.6479 0.7601										
433	9	7 8 9	0.897 0.0050 -0.0006	0.55 0.0000 0.0002	-0.00 0.0007 -0.0001	0.0940 -1.5669	-0.01 0.7024 0.8200	-0.96 0.0018 0.0012	0.05 -0.0000 -0.0001	-0.99 0.0002 0.0001	-0.0345 -0.4326	0.7030 0.6675										
433	10	7 8 9	0.897 0.0051 -0.0007	1.01 0.0001 0.0002	-0.00 0.0007 -0.0000	0.1017 -1.4632	-0.01 0.6851 0.6245	-0.95 0.0050 0.0041	0.05 -0.0002 0.0000	-0.99 0.0007 0.0006	0.2595 0.1133	0.7350 0.7309										
433	11	7 8 9	0.900 0.0050 -0.0006	1.02 0.0001 0.0002	-0.00 0.0007 -0.0000	0.1022 -1.8139	-0.01 0.6994 0.7992	-0.95 0.0051 0.0041	0.05 -0.0002 0.0000	-0.99 0.0007 0.0005	0.2498 0.1132	0.7217 0.7202										
433	12	7 8 9	0.898 0.0043 -0.0009	3.11 0.0000 0.0002	-0.00 0.0006 -0.0001	0.0918 -1.4657	-0.01 0.6994 0.7864	-0.93 0.0277 0.0269	0.05 -0.0006 0.0004	-0.97 0.0037 0.0035	0.1234 0.0877	0.6773 0.6653										
433	13	7 8 9	0.898 0.0043 -0.0009	3.11 0.0000 0.0002	-0.00 0.0006 -0.0001	0.0912 -1.4657	-0.02 0.6884 0.7864	-0.92 0.0278 0.0268	0.05 -0.0006 0.0004	-0.97 0.0037 0.0035	0.1220 0.0892	0.6751 0.6698										
433	14	7 8 9	0.899 0.0036 -0.0004	6.18 0.0000 0.0003	-0.00 0.0005 -0.0000	-0.0667 -3.3458	-0.01 0.7136 0.4713	-0.88 0.0590 0.0621	0.05 -0.0016 0.0013	-0.94 0.0076 0.0080	0.1377 0.1068	0.6464 0.6509										
433	15	7 8 9	0.898 0.0036 -0.0003	9.28 0.0000 0.0002	-0.00 0.0004 -0.0000	-0.1118 -3.4261	-0.02 0.6839 0.1199	-0.85 0.0842 0.0869	0.06 -0.0022 0.0019	-0.90 0.0101 0.0104	0.1363 0.1100	0.6048 0.6011										
433	16	7 8 9	0.898 0.0035 -0.0002	9.28 0.0000 0.0002	-0.00 0.0005 -0.0000	-0.1275 -5.5355	-0.01 0.7140 0.5194	-0.85 0.0840 0.0870	0.06 -0.0023 0.0019	-0.90 0.0102 0.0104	0.1377 0.1095	0.6082 0.5998										

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																			
433	17	0.897	9.28	-0.0004	-0.1260	-0.091	-0.85	0.06	-0.91	0.1364	0.6066	0.0036	-0.0004	-0.0004	-0.1260	-0.091	-0.85	0.06	-0.91	0.1364	0.6066
	6	-0.0004	0.0002	0.0000	-2.8157	-0.0944	0.0872	0.0019	0.0102	0.1092	0.6021	-0.0004	0.0000	0.0000	-2.8157	-0.0944	0.0872	0.0019	0.0102	0.1092	0.6021
433	18	0.899	9.28	0.0005	-0.1257	0.0630	-0.85	0.06	-0.91	0.1361	0.6063	0.0036	-0.0005	-0.0005	-0.1257	0.0630	-0.85	0.06	-0.91	0.1361	0.6063
	6	-0.0002	0.0002	-0.0000	-4.9166	0.0630	0.0874	0.0018	0.0104	0.1078	0.5992	-0.0002	0.0000	-0.0000	-4.9166	0.0630	0.0874	0.0018	0.0104	0.1078	0.5992
433	19	0.897	12.43	-0.0004	-0.1347	0.0865	-0.82	0.05	-0.89	0.1202	0.5930	0.0035	-0.0004	-0.0004	-0.1347	0.0865	-0.82	0.05	-0.89	0.1202	0.5930
	6	-0.0012	0.0002	-0.0001	-0.8796	0.4895	0.1076	0.0021	0.0125	0.0995	0.5831	-0.0012	-0.0001	-0.0001	-0.8796	0.4895	0.1076	0.0021	0.0125	0.0995	0.5831
433	20	0.897	12.42	0.0004	-0.1412	0.0857	-0.82	0.05	-0.89	0.1189	0.5944	0.0034	-0.0004	-0.0004	-0.1412	0.0857	-0.82	0.05	-0.89	0.1189	0.5944
	6	-0.0014	0.0002	-0.0000	-0.8009	0.2991	0.1082	0.0021	0.0126	0.0984	0.5852	-0.0014	-0.0000	-0.0000	-0.8009	0.2991	0.1082	0.0021	0.0126	0.0984	0.5852
433	21	0.899	12.43	0.0004	-0.1378	0.0893	-0.82	0.05	-0.89	0.1189	0.5946	0.0034	-0.0004	-0.0004	-0.1378	0.0893	-0.82	0.05	-0.89	0.1189	0.5946
	6	-0.0011	0.0001	-0.0001	-0.8797	0.6469	0.1080	0.0021	0.0126	0.0981	0.5853	-0.0011	-0.0001	-0.0001	-0.8797	0.6469	0.1080	0.0021	0.0126	0.0981	0.5853
433	22	0.897	-0.05	-0.0006	0.0433	-0.01	-0.96	0.05	-1.01	0.5344	1.1174	0.0051	-0.0006	-0.0006	0.0433	-0.01	-0.96	0.05	-1.01	0.5344	1.1174
	6	-0.0012	0.0002	-0.0000	-1.1569	0.2181	-0.0024	-0.0002	-0.0005	0.5700	1.0477	-0.0012	-0.0000	-0.0000	-1.1569	0.2181	-0.0024	-0.0002	-0.0005	0.5700	1.0477
433	23	0.897	-0.05	0.0006	0.0374	-0.02	-0.97	0.05	-1.00	0.5362	1.1027	0.0052	-0.0006	-0.0006	0.0374	-0.02	-0.97	0.05	-1.00	0.5362	1.1027
	6	-0.0010	0.0002	-0.0000	-1.3066	0.2869	-0.0025	-0.0002	-0.0005	0.5698	1.0299	-0.0010	-0.0000	-0.0000	-1.3066	0.2869	-0.0025	-0.0002	-0.0005	0.5698	1.0299
434	1	0.898	-3.08	-0.0006	-0.0991	-0.01	-0.11	0.05	-0.06	0.1450	0.6788	0.0042	-0.0006	-0.0006	-0.0991	-0.01	-0.11	0.05	-0.06	0.1450	0.6788
	6	-0.0015	0.0001	-0.0001	-0.6175	0.5045	-0.0277	-0.0007	-0.0037	0.1378	0.6697	-0.0015	-0.0001	-0.0001	-0.6175	0.5045	-0.0277	-0.0007	-0.0037	0.1378	0.6697
434	2	0.898	-1.08	0.0006	0.0176	-0.01	-0.08	0.05	-0.03	0.2473	0.7521	0.0047	-0.0006	-0.0006	0.0176	-0.01	-0.08	0.05	-0.03	0.2473	0.7521
	6	-0.0008	0.0002	-0.0001	-1.2052	0.5929	-0.0056	-0.0002	-0.0008	0.2484	0.7660	-0.0008	-0.0001	-0.0001	-1.2052	0.5929	-0.0056	-0.0002	-0.0008	0.2484	0.7660
434	3	0.898	-0.02	0.0006	0.0429	-0.01	-0.07	0.05	-0.03	0.0439	0.7033	0.0051	-0.0006	-0.0006	0.0429	-0.01	-0.07	0.05	-0.03	0.0439	0.7033
	6	-0.0007	0.0002	-0.0001	-1.4445	0.7352	-0.0022	-0.0000	-0.0003	-0.0213	0.7712	-0.0007	-0.0001	-0.0001	-1.4445	0.7352	-0.0022	-0.0000	-0.0003	-0.0213	0.7712
434	4	0.898	0.53	-0.0006	0.0815	-0.01	-0.06	0.05	-0.02	0.2560	0.7519	0.0049	-0.0006	-0.0006	0.0815	-0.01	-0.06	0.05	-0.02	0.2560	0.7519
	6	-0.0007	0.0002	-0.0001	-1.5640	0.7347	-0.0058	0.0001	0.0008	0.1563	0.7433	-0.0007	-0.0001	-0.0001	-1.5640	0.7347	-0.0058	0.0001	0.0008	0.1563	0.7433

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																			
434	5	7	0.896	0.001	-0.00	0.1089	-0.01	-0.05	0.05	-0.03	0.1246	0.7258	0.0050	0.0001	0.0007	0.13544	0.7065	0.0118	0.0002	0.0017	0.0792	0.7053
434	6	7	0.896	0.0002	-0.0000	0.0804	-0.01	-0.02	0.05	0.00	0.1267	0.6800	-0.0009	0.0002	0.0000	-1.4876	0.6742	0.0349	0.0003	0.0047	0.0964	0.6658
434	7	7	0.897	0.0003	-0.0005	-0.0889	-0.01	0.01	0.06	0.04	0.1418	0.6335	-0.0005	0.0003	0.0000	-2.9389	0.6223	0.0642	0.0018	0.0085	0.1078	0.6286
434	8	7	0.898	0.0002	-0.0005	-0.1071	-0.01	0.05	0.06	0.06	0.1401	0.6035	-0.0004	0.0000	0.0000	-2.9470	0.7121	0.0872	0.0024	0.0105	0.1139	0.5964
434	9	7	0.898	0.0002	-0.0005	-0.0867	-0.01	0.06	0.05	0.05	0.1159	0.5892	-0.0009	0.0002	-0.0000	-1.2361	0.7470	0.1095	0.0025	0.0129	0.0956	0.5832
434	10	7	0.898	0.0002	-0.0005	-0.0618	-0.01	-0.07	0.05	-0.03	0.0306	0.6276	-0.0011	0.0000	-0.0000	-1.2293	0.7228	0.0024	-0.0000	0.0002	-0.0898	0.5986
435	1	7	0.898	0.0001	-0.0006	-0.0944	-0.01	0.97	0.05	0.92	0.1471	0.6807	-0.0015	0.0001	-0.0001	-0.6254	0.4466	-0.0221	-0.0006	0.0028	0.1491	0.6875
435	2	7	0.898	0.0002	-0.0006	0.0076	-0.01	1.00	0.05	0.94	-3.9416	1.5452	-0.0008	0.0002	-0.0001	-1.2121	0.7096	0.0002	-0.0001	0.0000	-3.6572	0.7917
435	3	7	0.896	0.0003	-0.0006	0.0640	-0.01	1.02	0.05	0.96	0.2204	0.7708	-0.0006	0.0002	-0.0001	0.0640	0.7157	0.0073	0.0003	0.0011	0.1260	0.7202
435	4	7	0.895	0.0002	-0.0007	0.0915	-0.01	1.02	0.05	0.96	0.1208	0.7383	-0.0007	0.0002	-0.0001	-1.5420	0.8117	0.0138	0.0002	0.0019	0.0809	0.7033
435	5	7	0.896	0.0002	-0.0006	0.0960	-0.01	1.03	0.05	0.97	0.1118	0.7051	-0.0009	0.0002	-0.0001	0.0960	0.6840	0.0198	0.0004	0.0025	0.0769	0.6719

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																			
435	6	0.895	5.07	-0.00	0.0817	-0.01	1.06	0.05	0.99	0.1327	0.6730	0.0043	0.0000	0.0006	0.0017	0.0066	0.0427	0.0011	0.0057	0.1010	0.6654
	6	-0.0009	0.0002	-0.0001	-1.3931	0.8925	0.0415	0.0008	0.0055												
435	7	0.896	6.18	-0.00	0.0927	-0.01	1.10	0.06	1.03	0.1450	0.6235	0.0037	-0.0000	0.0005	0.0917	0.0696	0.0020	0.0086	0.1164	0.6152	
	6	-0.0006	0.0002	-0.0000	-2.4105	0.3482	0.0723	0.0016	0.0089												
435	8	0.897	9.30	-0.00	0.1384	-0.01	1.13	0.05	1.05	0.1356	0.5977	0.0035	-0.0000	0.0004	0.0699	0.0916	0.0024	0.0109	0.1140	0.5899	
	6	-0.0006	0.0002	-0.0000	-2.1710	0.1565	0.0935	0.0021	0.0110												
435	9	0.895	12.44	-0.00	0.1378	-0.01	1.15	0.05	1.06	0.1146	0.5862	0.0033	-0.0000	0.0004	0.6893	0.1118	0.0025	0.0131	0.0923	0.5776	
	6	-0.0011	0.0002	-0.0001	-0.9256	0.6611	0.1141	0.0021	0.0131												
435	10	0.898	-0.01	-0.00	0.0681	-0.01	1.01	0.05	0.96	0.1963	0.7159	0.0047	-0.0000	0.0007	0.7358	0.0075	0.0002	0.0310	0.1148	0.6660	
	6	-0.0010	0.0002	-0.0000	-1.3148	0.4311	0.0078	0.0001	0.0310												
436	6	0.898	-3.06	-0.00	0.0830	-0.02	2.95	0.05	3.03	0.2227	0.7020	0.0040	-0.0000	0.0005	0.7346	0.0076	0.0003	0.0010	0.2227	0.6950	
	6	-0.0010	0.0000	-0.0001	-0.4133	0.8231	-0.0055	-0.0002	-0.0007												
436	7	0.898	-1.03	-0.00	0.0254	-0.02	2.98	0.05	3.05	0.1662	0.7853	0.0045	-0.0001	0.0006	0.6777	0.0105	0.0003	0.0016	0.0793	0.7425	
	6	-0.0003	0.0001	-0.0001	-1.6898	2.1423	0.0130	0.0002	0.0019												
436	8	0.899	-0.00	-0.00	0.1003	-0.01	2.99	0.05	3.06	0.1205	0.7143	0.0046	-0.0001	0.0007	0.7574	0.0226	0.0005	0.0032	0.0807	0.6813	
	6	-0.0004	0.0001	-0.0001	-1.8827	1.6097	0.0239	0.0003	0.0032												
436	9	0.901	0.53	-0.00	0.0665	-0.01	3.00	0.05	3.07	0.1219	0.7000	0.0051	-0.0001	0.0006	0.6747	0.0284	0.0005	0.0040	0.0866	0.6847	
	6	-0.0004	0.0001	-0.0001	-2.1213	1.4137	0.0292	0.0005	0.0040												
436	10	0.899	1.02	-0.00	0.0724	-0.01	3.01	0.05	3.07	0.1241	0.6927	0.0052	-0.0001	0.0007	0.6759	0.0340	0.0008	0.0047	0.0876	0.6779	
	6	-0.0004	0.0002	-0.0001	-2.3158	1.7172	0.0344	0.0006	0.0046												
436	11	0.901	3.10	-0.00	0.0188	-0.02	3.04	0.05	3.10	0.1416	0.6582	0.0047	-0.0000	0.0006	0.6834	0.0550	0.0015	0.0072	0.1031	0.6549	
	6	-0.0007	0.0002	-0.0001	-1.6593	1.2017	0.0562	0.0011	0.0073												

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																													
436	12	7	0.901	6.21	-0.00	-0.0927	-0.02	3.07	0.06	3.12	0.1478	0.6139	6	0.0037	0.0000	0.0005	0.6917	0.784	0.0023	0.0096	9	-0.0001	0.0002	-0.0000	-13.1168	4.3495	0.0821	0.0019	0.0099	0.1170	0.6044
436	13	7	0.901	9.28	-0.00	-0.1347	-0.02	3.09	0.05	3.14	0.1296	0.5946	6	0.0035	0.0000	0.0004	0.6865	0.999	0.0025	0.0118	9	-0.0002	0.0002	-0.0000	-3.6913	1.0866	0.1024	0.0021	0.0119	0.1040	0.5839
436	14	7	0.900	12.47	-0.00	-0.1283	-0.02	3.11	0.05	3.15	0.1050	0.5866	6	0.0035	0.0000	0.0004	0.6873	0.1167	0.0024	0.0136	9	-0.0007	0.0001	-0.0001	-1.1724	0.6498	0.1212	0.0019	0.0139	0.0791	0.5764
436	15	7	0.900	0.00	-0.00	0.0663	-0.02	2.99	0.05	3.06	0.1142	0.7031	6	0.0048	0.0000	0.0006	0.6963	0.222	0.0005	0.0031	9	-0.0007	0.0002	-0.0001	-1.4603	0.9061	0.0236	0.0003	0.0031	0.0749	0.6609
437	1	7	0.899	-3.33	-0.00	-0.0501	-0.02	6.01	0.05	6.01	0.1470	0.7914	6	0.0039	0.0000	0.0005	0.7239	0.0102	0.0003	0.0016	9	-0.0011	0.0001	-0.0001	-0.5900	0.6888	0.0127	0.0001	0.0018	0.0697	0.7232
437	2	7	0.901	-1.01	-0.00	0.0434	-0.01	6.04	0.05	6.03	0.1173	0.7019	6	0.0045	0.0000	0.0006	0.7133	0.324	0.0007	0.0045	9	-0.0008	0.0002	-0.0001	-1.2013	0.8418	0.0350	0.0005	0.0047	0.0846	0.6814
437	3	7	0.900	0.00	-0.00	0.0529	-0.02	6.06	0.05	6.05	0.1318	0.6870	6	0.0049	0.0000	0.0007	0.7193	0.0435	0.0011	0.0059	9	-0.0009	0.0002	-0.0001	-1.3018	0.8684	0.0453	0.0008	0.0060	0.0960	0.6705
437	4	7	0.899	0.55	-0.00	0.0319	-0.01	6.06	0.05	6.06	0.1372	0.6748	6	0.0054	0.0000	0.0007	0.6712	0.0491	0.0013	0.0066	9	-0.0008	0.0002	-0.0001	-1.5014	0.9537	0.0506	0.0010	0.0067	0.0999	0.6632
437	5	7	0.898	1.07	-0.00	0.0462	-0.01	6.08	0.05	6.05	0.1425	0.6636	6	0.0054	0.0000	0.0007	0.6838	0.340	0.0015	0.0071	9	-0.0009	0.0002	-0.0001	-1.3551	0.8507	0.0555	0.0011	0.0072	0.1028	0.6565
437	6	7	0.898	3.15	-0.00	-0.0045	-0.01	6.10	0.06	6.08	0.1542	0.6196	6	0.0050	0.0000	0.0006	0.6640	0.0705	0.0021	0.0087	9	-0.0011	0.0002	-0.0001	-1.2016	0.6720	0.0722	0.0016	0.0089	0.1160	0.6177
437	7	7	0.899	6.24	-0.00	-0.1079	-0.02	6.13	0.06	6.11	0.1373	0.5970	6	0.0037	0.0000	0.0005	0.6870	0.0914	0.0025	0.0109	9	-0.0004	0.0002	-0.0000	-3.4865	0.7317	0.0937	0.0021	0.0110	0.1167	0.5884

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key													
437	8	0.904	9.30	-0.0000	-0.0004	-0.1388	-0.02	6.14	0.025	6.12	0.1192	0.5838			
	8	0.0035	-0.0000	0.0004	-0.0000	-0.9729	0.6568	0.1079	0.0020	0.0127	0.0936	0.5743			
	9	-0.0006	0.0002	-0.0000	-0.0000	-1.9729	0.3536	0.1112	0.0020	0.0127	0.0936	0.5743			
437	9	0.900	12.45	-0.0000	-0.0005	-0.1113	-0.02	6.15	0.021	6.11	0.0672	0.5822			
	8	0.0035	-0.0000	0.0005	-0.0001	-0.8280	0.7192	0.1206	0.0021	0.0140	0.0654	0.5720			
	9	-0.0010	0.0001	-0.0001	-0.0001	-0.8280	0.8028	0.1213	0.0015	0.0138	0.0654	0.5720			
437	10	0.901	0.00	-0.0000	-0.0001	0.0371	-0.01	6.06	0.06	6.05	0.1278	0.6796			
	8	0.0048	0.0000	0.0006	-0.0001	-1.0765	0.7056	0.0435	0.0011	0.0059	0.0936	0.6626			
	9	-0.0013	0.0002	-0.0001	-0.0001	-1.0765	0.4500	0.0435	0.0008	0.0060	0.0936	0.6626			
438	1	0.901	-3.01	-0.0000	-0.0004	-0.0577	-0.01	9.08	0.007	9.01	0.1133	0.7133			
	8	0.0036	-0.0000	0.0004	-0.0001	-0.5609	0.6791	0.0318	0.0005	0.0048	0.0818	0.6912			
	9	-0.0013	0.0001	-0.0001	-0.0001	-0.5609	0.6631	0.0347	0.0005	0.0048	0.0818	0.6912			
438	2	0.902	-1.00	-0.0000	-0.0002	-0.0313	-0.01	9.11	0.005	9.04	0.1377	0.6661			
	8	0.0045	0.0000	0.0006	-0.0002	-1.3277	0.7301	0.0520	0.0014	0.0069	0.1025	0.6542			
	9	-0.0008	0.0002	-0.0002	-0.0002	-1.3277	1.2157	0.0548	0.0011	0.0071	0.1025	0.6542			
438	3	0.901	0.03	-0.0000	-0.0007	0.0118	-0.01	9.13	0.005	9.04	0.1477	0.6425			
	8	0.0053	0.0000	0.0007	-0.0002	-1.3845	0.6749	0.0608	0.0017	0.0078	0.1093	0.6385			
	9	-0.0008	0.0002	-0.0002	-0.0002	-1.3845	1.1923	0.0630	0.0013	0.0080	0.1093	0.6385			
438	4	0.897	0.57	-0.0000	-0.0001	-0.0048	-0.01	9.14	0.006	9.05	0.1487	0.6306			
	8	0.0056	0.0000	0.0007	-0.0001	-1.4191	0.6617	0.0655	0.0019	0.0084	0.1115	0.6284			
	9	-0.0009	0.0002	-0.0001	-0.0001	-1.4191	1.0104	0.0673	0.0015	0.0084	0.1115	0.6284			
438	5	0.899	1.06	-0.0000	-0.0007	0.0113	-0.01	9.14	0.005	9.05	0.1544	0.6240			
	8	0.0057	0.0000	0.0007	-0.0001	-1.1894	0.6712	0.0687	0.0021	0.0085	0.1143	0.6182			
	9	-0.0012	0.0002	-0.0001	-0.0001	-1.1894	0.5922	0.0707	0.0016	0.0087	0.1143	0.6182			
438	6	0.900	3.17	-0.0000	-0.0006	-0.0132	-0.01	9.17	0.005	9.08	0.1485	0.6045			
	8	0.0051	-0.0000	0.0006	-0.0001	-1.4632	0.6600	0.0840	0.0024	0.0101	0.1240	0.5977			
	9	-0.0009	0.0002	-0.0001	-0.0001	-1.4632	1.0195	0.0846	0.0020	0.0101	0.1240	0.5977			
438	7	0.899	6.22	-0.0000	-0.0005	-0.1013	-0.01	9.18	0.006	9.09	0.1251	0.5892			
	8	0.0039	-0.0000	0.0005	-0.0000	-3.5982	0.6611	0.1037	0.0025	0.0122	0.1000	0.5766			
	9	-0.0004	0.0003	-0.0000	-0.0000	-3.5982	0.6662	0.1053	0.0021	0.0122	0.1000	0.5766			
438	8	0.901	9.31	-0.0000	-0.0004	-0.1351	-0.02	9.19	0.005	9.08	0.0982	0.5827			
	8	0.0035	-0.0000	0.0004	-0.0000	-1.5956	0.6730	0.1149	0.0022	0.0135	0.0747	0.5752			
	9	-0.0007	0.0002	-0.0000	-0.0000	-1.5956	0.2775	0.1180	0.0017	0.0135	0.0747	0.5752			

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																				
438	9	7	0.900	-12.44	-0.00	-0.1065	-0.02	9.19	0.05	9.08	0.0813	0.5801	0.0035	-0.0000	0.0000	0.0004	0.6798	0.1211	0.0019	0.0140	0.0614	0.5702	
		6	-0.0012	0.0002	-0.0001	-0.9683	0.4442	0.1246	0.0015	0.0142													
438	10	7	0.899	0.02	-0.00	0.0102	-0.01	9.13	0.06	9.04	0.1426	0.6360	0.0054	0.0000	0.0007	0.0000	0.6648	0.0613	0.0017	0.0077	0.1079	0.6359	
		6	-0.0012	0.0003	-0.0001	-1.2267	0.6064	0.0631	0.0013	0.0080													
439	1	7	0.899	-2.99	-0.00	-0.0315	-0.01	12.05	0.05	12.06	0.1316	0.6719	0.0037	-0.0000	0.0005	-0.0000	0.6855	0.0511	0.0013	0.0068	0.1026	0.6596	
		6	-0.0012	0.0001	-0.0001	-0.7004	0.7300	0.0547	0.0011	0.0072													
439	2	7	0.898	-0.97	-0.00	0.0084	-0.01	12.08	0.05	12.08	0.1505	0.6312	0.0047	0.0000	0.0006	0.0000	0.7295	0.0675	0.0020	0.0085	0.1106	0.6187	
		6	-0.0013	0.0002	-0.0001	-0.9991	0.5837	0.0712	0.0015	0.0088													
439	3	7	0.900	0.04	-0.00	-0.0004	-0.01	12.09	0.05	12.09	0.1571	0.6158	0.0055	-0.0000	0.0007	-0.0000	0.6858	0.0743	0.0023	0.0091	0.1179	0.6105	
		6	-0.0009	0.0002	-0.0001	-1.4676	0.9320	0.0774	0.0018	0.0094													
439	4	7	0.900	0.59	-0.00	0.0041	-0.01	12.10	0.05	12.09	0.1556	0.6196	0.0057	0.0000	0.0007	-0.0001	0.6688	0.0781	0.0024	0.0095	0.1212	0.6038	
		6	-0.0009	0.0002	-0.0001	-1.3931	0.8925	0.0806	0.0019	0.0097													
439	5	7	0.900	1.09	-0.00	-0.0074	-0.01	12.10	0.05	12.10	0.1510	0.6063	0.0059	-0.0000	0.0007	-0.0001	0.6486	0.0821	0.0024	0.0100	0.1261	0.6009	
		6	-0.0009	0.0002	-0.0001	-1.4504	0.8464	0.0833	0.0021	0.0100													
439	6	7	0.900	5.18	-0.00	-0.0102	-0.01	12.12	0.06	12.10	0.1348	0.5955	0.0049	-0.0000	0.0006	-0.0001	0.7006	0.0962	0.0025	0.0114	0.1097	0.5824	
		6	-0.0011	0.0003	-0.0001	-1.2929	0.7823	0.0967	0.0021	0.0112													
439	7	7	0.898	6.24	-0.00	-0.1155	-0.02	12.13	0.06	12.11	0.1136	0.5823	0.0039	-0.0000	0.0003	-0.0000	0.6687	0.1082	0.0024	0.0128	0.0883	0.5726	
		6	-0.0006	0.0003	-0.0000	-2.4331	0.4652	0.1119	0.0019	0.0128													
439	8	7	0.899	9.30	-0.00	-0.1087	-0.02	12.12	0.05	12.11	0.0873	0.5804	0.0037	-0.0000	0.0005	-0.0000	0.6619	0.1161	0.0020	0.0134	0.0649	0.5719	
		6	-0.0006	0.0002	-0.0000	-1.9529	0.4062	0.1189	0.0015	0.0136													
439	9	7	0.900	12.46	-0.00	-0.1185	-0.02	12.13	0.06	12.11	0.0822	0.5748	0.0038	-0.0000	0.0004	-0.0001	0.6459	0.1230	0.0020	0.0141	0.0616	0.5657	
		6	-0.0011	0.0002	-0.0001	-0.9238	0.5803	0.1252	0.0015	0.0141													

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key															
439	10	7	0.0041	-0.0000	-2.04	-0.0000	0.0005	-0.0047	-0.01	12.07	0.0017	0.05	12.07	0.0076	0.1432	0.6432		
		8	-0.0012	0.0002	-0.0002	-0.0001	-0.0001	-0.9581	0.6851	0.0593	0.0014	0.0014	0.0080	0.0080	0.1119	0.6359		
		9							0.7158	0.0629	0.0014	0.0014	0.0080	0.0080				
439	11	7	0.900	0.0000	0.06	-0.00	0.0007	0.0001	-0.01	12.09	0.0023	0.05	12.09	0.0090	0.1568	0.6115		
		8	0.0054	0.0002	0.0000	0.0007	0.0001	-1.2066	0.6788	0.0742	0.0018	0.0018	0.0093	0.0093	0.1183	0.6080		
		9	-0.0011	0.0002	-0.0002	-0.0001	-1.2066	0.8091	0.8091	0.0771	0.0018	0.0018	0.0093	0.0093				
440	1	7	0.897	-2.98	-0.0001	-0.00	-0.0415	-0.01	-0.01	15.09	0.0015	0.05	15.06	0.0083	0.1458	0.6284		
		8	0.0038	-0.0000	0.0000	0.0005	-0.7450	0.6914	0.7862	0.0666	0.0015	0.0015	0.0087	0.0087	0.1098	0.6206		
		9	-0.0013	0.0001	-0.0001	-0.0002	-0.7450	0.7862	0.7862	0.0707	0.0015	0.0015	0.0087	0.0087				
440	2	7	0.897	-0.95	-0.0002	-0.00	0.0023	-0.01	-0.01	15.10	0.0024	0.05	15.08	0.0098	0.1520	0.6064		
		8	0.0049	0.0000	0.0000	0.0007	0.0023	0.7234	0.7234	0.0810	0.0024	0.0024	0.0098	0.0098	0.1291	0.6006		
		9	-0.0012	0.0002	-0.0002	-0.0001	-1.0404	0.6933	0.6933	0.0830	0.0021	0.0021	0.0099	0.0099				
440	3	7	0.899	0.06	-0.0002	-0.00	-0.0267	-0.01	-0.01	15.12	0.0025	0.05	15.09	0.0107	0.1418	0.6029		
		8	0.0057	-0.0000	0.0000	0.0007	-0.3118	0.6360	0.6360	0.0891	0.0025	0.0025	0.0107	0.0107	0.1241	0.5882		
		9	-0.0009	0.0002	-0.0002	-0.0002	-1.3118	1.0475	1.0475	0.0890	0.0022	0.0022	0.0104	0.0104				
440	4	7	0.898	0.60	-0.0002	-0.00	-0.0173	-0.01	-0.01	15.12	0.0021	0.05	15.08	0.0108	0.1410	0.5977		
		8	0.0059	-0.0000	0.0000	0.0007	-0.0173	0.6456	0.6456	0.0916	0.0021	0.0021	0.0108	0.0108	0.1169	0.5848		
		9	-0.0013	0.0002	-0.0002	-0.0001	-1.0736	0.6099	0.6099	0.0925	0.0021	0.0021	0.0108	0.0108				
440	5	7	0.898	1.10	-0.0002	-0.00	-0.0176	-0.01	-0.01	15.12	0.0021	0.05	15.08	0.0111	0.1363	0.5912		
		8	0.0059	-0.0000	0.0000	0.0007	-0.0176	0.6377	0.6377	0.0948	0.0021	0.0021	0.0112	0.0112	0.1251	0.5798		
		9	-0.0011	0.0002	-0.0002	-0.0001	-1.1940	0.7222	0.7222	0.0950	0.0021	0.0021	0.0111	0.0111	0.1016	0.5742		
440	6	7	0.896	3.16	-0.0002	-0.00	-0.0121	-0.01	-0.01	15.13	0.0021	0.06	15.09	0.0119	0.0953	0.5800		
		8	0.0052	-0.0000	0.0000	0.0007	-0.0121	0.6872	0.6872	0.1028	0.0021	0.0021	0.0119	0.0119	0.0718	0.5719		
		9	-0.0011	0.0002	-0.0002	-0.0001	-1.3150	0.8484	0.8484	0.1054	0.0021	0.0021	0.0121	0.0121				
440	7	7	0.897	6.22	-0.0003	-0.00	-0.0951	-0.01	-0.01	15.13	0.0016	0.06	15.09	0.0131	0.0953	0.5800		
		8	0.0039	-0.0000	0.0000	0.0005	-0.0951	0.6860	0.6860	0.1120	0.0016	0.0016	0.0131	0.0131	0.0718	0.5719		
		9	-0.0004	0.0003	-0.0003	-0.0000	-3.4802	0.7639	0.7639	0.1145	0.0016	0.0016	0.0131	0.0131				
440	8	7	0.897	9.32	-0.0002	-0.00	-0.1019	-0.01	-0.01	15.13	0.0015	0.05	15.10	0.0135	0.0609	0.5750		
		8	0.0037	-0.0000	0.0000	0.0005	-0.1019	0.7001	0.7001	0.1180	0.0015	0.0015	0.0135	0.0135	0.0648	0.5704		
		9	-0.0005	0.0002	-0.0002	-0.0000	-2.2234	0.4016	0.4016	0.1211	0.0015	0.0015	0.0138	0.0138				
440	9	7	0.898	12.45	-0.0001	-0.00	-0.1009	-0.01	-0.01	15.14	0.0012	0.06	15.10	0.0144	0.0752	0.5683		
		8	0.0039	-0.0000	0.0000	0.0005	-0.1009	0.6732	0.6732	0.1249	0.0012	0.0012	0.0144	0.0144	0.0752	0.5683		
		9	-0.0012	0.0002	-0.0002	-0.0001	-0.9211	0.6310	0.6310	0.1268	0.0012	0.0012	0.0144	0.0144				

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key															
440	10	7	0.897	0.0000	-0.0007	-0.0126	-0.01	15.12	0.0025	0.06	15.09	0.1466	0.5946				
	8	0.0057	-0.0000	0.0007	-0.0427	0.6562	0.0675	0.0025	0.0104								
	9	-0.00015	0.00003	-0.00001	-1.0427	0.4482	0.0690	0.0022	0.0104								
441	1	7	0.799	-3.13	-0.00	-0.0028	-0.01	-3.03	0.05	-3.03	0.2632	0.6272					
	6	0.0036	-0.0000	0.0005	-0.6157	0.6815	-0.0431	-0.0022	-0.0054								
	9	-0.00020	0.00002	-0.00001	-0.6157	0.4660	-0.0405	-0.0020	-0.0052								
441	2	7	0.800	-1.09	-0.00	0.0614	-0.01	-2.99	0.05	-3.00	0.2803	0.6790					
	8	0.0046	0.0000	0.0006	0.9700	0.7032	-0.0229	-0.0012	-0.0031								
	9	-0.00013	0.00002	-0.00001	-0.9700	0.5436	-0.0198	-0.0012	-0.0026								
441	3	7	0.799	-0.06	-0.00	0.0616	-0.01	-2.97	0.06	-2.98	0.2850	0.7118					
	8	0.0053	0.0000	0.0006	-1.2146	0.6442	-0.0132	-0.0007	-0.0018								
	9	-0.00011	0.00002	-0.00001	-1.2146	0.4852	-0.0108	-0.0007	-0.0015								
441	4	7	0.799	0.49	-0.00	0.0731	-0.01	-2.96	0.06	-2.98	0.3227	0.7864					
	8	0.0056	0.0000	0.0007	-1.3331	0.6334	-0.0080	-0.0005	-0.0012								
	9	-0.00010	0.00002	-0.00001	-1.3331	0.4804	-0.0067	-0.0005	-0.0010								
441	5	7	0.800	1.02	-0.00	0.0936	-0.01	-2.95	0.06	-2.97	0.3323	0.9036					
	8	0.0054	0.0001	0.0007	0.0936	0.6462	-0.0037	-0.0002	-0.0006								
	9	-0.00011	0.00002	-0.00001	-1.2213	0.4399	-0.0028	-0.0003	-0.0004								
441	6	7	0.799	3.07	-0.00	0.0878	-0.01	-2.91	0.06	-2.94	0.2383	0.6132					
	8	0.0048	0.0000	0.0006	0.0878	0.6524	0.0126	0.0006	0.0015								
	9	-0.00013	0.00003	-0.00001	-1.1449	0.3857	0.0112	0.0004	0.0014								
441	9	7	0.799	6.15	-0.00	0.0099	-0.04	-2.85	0.06	-2.87	0.2492	0.6281					
	6	0.0034	0.0000	0.0004	0.0099	0.7009	0.0393	0.0019	0.0049								
	9	-0.00000	0.00002	-0.00000	-16.4590	2.4371	0.0423	0.0017	0.0053								
441	10	7	0.799	9.29	-0.00	0.0008	-0.03	-2.80	0.06	-2.83	0.2589	0.5746					
	8	0.0033	0.0000	0.0004	0.0008	0.6881	0.0624	0.0032	0.0071								
	9	-0.00000	0.00002	-0.00000	-14.7803	-1.5388	0.0668	0.0028	0.0077								
441	11	7	0.799	12.43	-0.00	0.0322	-0.04	-2.78	0.06	-2.83	0.1925	0.5523					
	8	0.0031	-0.0000	0.0004	-0.0322	0.6707	0.0795	0.0030	0.0092								
	9	-0.00008	0.00001	-0.00001	-1.0827	0.7464	0.0836	0.0026	0.0092								
441	12	7	0.800	-0.09	-0.00	0.0579	-0.02	-2.97	0.05	-2.97	0.2594	0.6293					
	8	0.0052	0.0000	0.0007	0.0579	0.6727	-0.0136	-0.0007	-0.0019								
	9	-0.00008	0.00002	-0.00001	-1.3479	0.6878	-0.0109	-0.0007	-0.0015								

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																																																																																																							
		7 8 9	-3.10 0.0000 0.0001	-0.00 0.0005 -0.0002	0.0016 -0.6725 0.0001	0.0016 -0.6725 0.0001	-0.03 0.7564 0.0001	-1.02 -0.0315 -0.0282	0.06 -0.0016 -0.0015	-1.06 -0.0040 -0.0036	7 8 9	0.801 0.0046 -0.0009	-0.00 0.0006 -0.0001	0.0516 -1.1137 0.0001	0.0516 -1.1137 0.0001	-0.03 0.7168 0.8618	-0.97 -0.0111 -0.0085	0.06 -0.0005 -0.0006	-1.02 -0.0015 -0.0011	7 8 9	0.800 0.0054 -0.0007	-0.00 0.0000 0.0002	0.0644 -1.4563 0.0001	0.0644 -1.4563 0.0001	-0.03 0.6448 1.1116	-0.95 -0.0025 -0.0012	0.06 -0.0001 -0.0002	-1.01 -0.0004 -0.0002	7 8 9	0.800 0.0055 -0.0007	0.00 0.0000 0.0002	0.0602 -1.5204 0.0001	0.0602 -1.5204 0.0001	-0.03 0.6435 1.0597	-0.95 -0.0019 -0.0010	0.05 -0.0001 -0.0002	-1.01 -0.0003 -0.0001	7 8 9	0.799 0.0053 -0.0006	1.00 0.0001 0.0002	-0.00 0.0007 -0.0001	0.0998 -1.6822 0.0001	0.0998 -1.6822 0.0001	-0.02 0.6766 1.2722	-0.94 0.0049 0.0052	0.05 -0.0003 0.0001	-0.99 0.0006 0.0006	7 8 9	0.801 0.0046 -0.0008	3.08 0.0000 0.0002	-0.00 0.0006 -0.0001	0.0983 -1.2852 0.0001	0.0983 -1.2852 0.0001	-0.03 0.7052 0.7847	-0.90 0.0250 0.0230	0.06 -0.0012 0.0009	-0.96 0.0031 0.0029	7 8 9	0.799 0.0035 -0.0000	6.18 0.0000 0.0002	-0.00 0.0004 -0.0000	0.0263 -21.3386 0.0001	0.0263 -21.3386 0.0001	-0.03 0.7030 2.8217	-0.84 0.0505 0.0533	0.06 -0.0025 0.0022	-0.91 0.0061 0.0064	7 8 9	0.798 0.0030 -0.0002	9.28 0.0000 0.0002	-0.00 0.0004 -0.0000	0.0278 -5.3854 0.0001	0.0278 -5.3854 0.0001	-0.02 0.7523 0.4564	-0.81 0.0717 0.0755	0.06 -0.0033 0.0028	-0.88 0.0081 0.0085	7 8 9	0.797 0.0033 -0.0008	12.46 0.0000 0.0002	-0.00 0.0004 -0.0001	0.0110 -1.4454 0.0001	0.0110 -1.4454 0.0001	-0.02 0.6919 0.6515	-0.80 0.0852 0.0876	0.05 -0.0026 0.0022	-0.88 0.0094 0.0097	7 8 9	0.800 0.0053 -0.0007	0.00 0.0000 0.0002	-0.00 0.0007 -0.0001	0.0622 -1.6071 0.0001	0.0622 -1.6071 0.0001	-0.02 0.6590 1.0762	-0.96 -0.0019 -0.0009	0.06 -0.0001 -0.0002	-1.01 -0.0004 -0.0002	7 8 9	0.798 0.0035 -0.0015	-3.10 0.0000 -0.0002	-0.00 0.0004 -0.0002	-0.0096 -0.7139 0.0001	-0.0096 -0.7139 0.0001	-0.02 0.6934 0.6546	-0.13 -0.0256 -0.0216

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key									
443	2	0.799	-1.05	-0.00	0.0504	-0.02	-0.09	0.05	-0.03	0.2901	0.7350
	6	0.0046	0.0000	0.0006	0.0504	0.6916	-0.0059	-0.0003	-0.0008	0.6061	0.7367
	9	-0.0008	0.0002	-0.0001	-1.3176	1.0212	-0.0031	-0.0003	-0.0004		
443	3	0.800	-0.05	-0.00	0.0770	-0.02	-0.06	0.06	-0.03	0.2867	0.5859
	6	0.0053	0.0000	0.0007	0.0770	0.6775	0.0017	0.0000	0.0002	-0.0311	0.6679
	9	-0.0006	0.0002	-0.0001	-1.7817	1.4475	0.0026	-0.0000	0.0003		
443	4	0.799	0.00	-0.00	0.0671	-0.02	-0.06	0.06	-0.02	0.2405	0.5234
	6	0.0052	0.0000	0.0007	0.0671	0.6680	0.0023	0.0001	0.0002	0.0477	0.7005
	9	-0.0007	0.0002	-0.0001	-1.5113	1.0896	0.0031	0.0000	0.0004		
443	5	0.800	1.01	-0.00	0.0755	-0.02	-0.05	0.06	-0.00	0.2663	0.6762
	6	0.0054	0.0000	0.0007	0.0755	0.6410	0.0099	0.0005	0.0013	0.1791	0.6798
	9	-0.0006	0.0002	-0.0001	-1.6358	1.1159	0.0103	0.0003	0.0014		
443	6	0.800	3.12	-0.00	0.0971	-0.02	-0.00	0.05	0.02	0.2490	0.6299
	6	0.0046	0.0000	0.0006	0.0971	0.6801	0.0311	0.0015	0.0039	0.2023	0.6411
	9	-0.0006	0.0002	-0.0001	-1.7411	1.2450	0.0297	0.0012	0.0036		
443	7	0.799	6.19	-0.00	0.0278	-0.02	0.05	0.06	0.07	0.2582	0.5985
	6	0.0034	0.0000	0.0004	0.0278	0.6944	0.0547	0.0028	0.0065	0.2137	0.6012
	9	-0.0001	0.0002	-0.0000	8.8929	-1.0879	0.0580	0.0024	0.0069		
443	8	0.799	9.28	-0.00	0.0328	-0.02	0.07	0.06	0.09	0.2198	0.5624
	6	0.0031	0.0000	0.0004	0.0328	0.7651	0.0757	0.0033	0.0085	0.1853	0.5633
	9	-0.0000	0.0002	-0.0000	-18.4366	2.2992	0.0786	0.0029	0.0088		
443	9	0.799	12.47	-0.00	0.0436	-0.02	0.07	0.06	0.09	0.1462	0.5553
	6	0.0034	0.0000	0.0005	0.0436	0.7331	0.0860	0.0025	0.0095	0.1151	0.5574
	9	-0.0004	0.0002	-0.0000	-2.6666	0.7413	0.0886	0.0020	0.0098		
443	10	0.799	-0.02	-0.00	0.0750	-0.02	-0.06	0.06	-0.02	0.2303	0.5196
	6	0.0053	0.0000	0.0007	0.0750	0.6844	0.0021	0.0000	0.0002	-0.0229	0.5534
	9	-0.0007	0.0002	-0.0001	-1.5823	1.2644	0.0032	-0.0000	0.0003		
444	1	0.800	-3.08	-0.00	0.0060	-0.03	0.97	0.06	0.92	0.2796	0.6604
	6	0.0037	0.0000	0.0005	0.0060	0.7040	-0.0185	-0.0010	-0.0024	0.3264	0.6724
	9	-0.0014	0.0001	-0.0002	-0.6736	0.7895	-0.0154	-0.0010	-0.0020		
444	2	0.801	-1.04	-0.00	0.0551	-0.02	1.01	0.06	0.95	2.1050	2.4431
	6	0.0046	0.0000	0.0006	0.0551	0.7339	-0.0001	-0.0000	-0.0000	-0.4568	0.6102
	9	-0.0008	0.0002	-0.0001	-1.2307	1.0303	0.0011	-0.0001	-0.0001		

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key															
444	3	7	0.799	-0.02	-0.00	0.0629	-0.02	1.02	0.05	0.97	0.3146	0.6857						
		8	0.0052	0.0000	0.0007	0.0629	0.6737	0.0066	0.0004	0.0009	0.1628	0.6823						
		9	-0.0006	0.0002	-0.0001	-1.6029	1.2072	0.0078	0.0002	0.0010								
444	4	7	0.799	0.01	-0.00	0.0603	-0.02	1.02	0.05	0.97	0.2956	0.6653						
		8	0.0053	0.0000	0.0007	0.0603	0.6580	0.0069	0.0004	0.0009	0.1640	0.6763						
		9	-0.0007	0.0002	-0.0001	-1.4632	1.0195	0.0081	0.0002	0.0010								
444	5	7	0.800	1.00	-0.00	0.0998	-0.02	1.04	0.05	0.98	0.2435	0.6486						
		8	0.0053	0.0001	0.0007	0.0998	0.6766	0.0167	0.0008	0.0021	0.1994	0.6439						
		9	-0.0006	0.0002	-0.0001	-1.8565	1.3863	0.0163	0.0006	0.0021								
444	6	7	0.798	3.10	-0.00	0.0887	-0.02	1.08	0.06	1.02	0.2450	0.6300						
		8	0.0047	0.0000	0.0006	0.0887	0.6692	0.377	0.0018	0.0047	0.2047	0.6496						
		9	-0.0006	0.0002	-0.0001	-1.5925	1.2362	0.0361	0.0014	0.0046								
444	7	7	0.799	6.20	-0.00	0.0283	-0.02	1.14	0.06	1.06	0.2571	0.5792						
		8	0.0034	0.0000	0.0004	0.0283	0.7058	0.590	0.0030	0.0068	0.2099	0.5826						
		9	0.0001	0.0002	-0.0000	11.1299	-2.6434	0.0625	0.0026	0.0072								
444	8	7	0.801	9.35	-0.00	0.0309	-0.02	1.16	0.06	1.08	0.2106	0.5545						
		8	0.0033	0.0000	0.0004	0.0309	0.7204	0.783	0.0032	0.0086	0.1707	0.5565						
		9	0.0003	0.0002	-0.0000	4.1510	-0.6886	0.0811	0.0027	0.0090								
444	9	7	0.799	12.48	-0.00	0.0158	-0.02	1.15	0.06	1.07	0.1337	0.5551						
		8	0.0033	0.0000	0.0004	0.0158	0.7042	0.872	0.0023	0.0096	0.1061	0.5563						
		9	-0.0005	0.0002	-0.0000	-2.0610	0.8514	0.0903	0.0019	0.0100								
444	10	7	0.800	0.50	-0.00	0.0694	-0.02	1.03	0.06	0.97	0.2221	0.6349						
		8	0.0054	0.0000	0.0007	0.0694	0.6532	0.618	0.0005	0.0015	0.1842	0.6513						
		9	-0.0006	0.0002	-0.0001	-1.9711	1.3275	0.0118	0.0004	0.0015								
444	11	7	0.800	-0.03	-0.00	0.0651	-0.02	1.02	0.06	0.97	0.2504	0.6077						
		8	0.0054	0.0000	0.0007	0.0651	0.6592	0.070	0.0003	0.0008	0.1438	0.6337						
		9	-0.0006	0.0002	-0.0001	-1.8122	1.2825	0.0081	0.0002	0.0010								
445	1	7	0.799	-3.07	-0.00	0.0087	-0.03	2.95	0.05	3.04	0.3323	0.7508						
		8	0.0035	0.0000	0.0005	0.0087	0.7393	-0.066	-0.0004	-0.0010	0.3233	0.7528						
		9	-0.0010	0.0001	-0.0002	-0.8174	1.2599	-0.0035	-0.0004	-0.0005	0.5690	0.7328						
445	2	7	0.799	-1.00	-0.00	0.0555	-0.03	2.99	0.06	3.06	0.2351	0.6819						
		8	0.0046	0.0000	0.0006	0.0555	0.7423	0.0094	0.0004	0.0012	0.2351	0.6819						
		9	-0.0008	0.0002	-0.0001	-1.3704	1.1981	0.0116	0.0004	0.0015	0.1839	0.6707						

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key													
445	3	7	0.799	0.0000	-0.000	0.0660	-0.03	0.6977	0.0190	0.06	3.08	0.2389	0.6511			
		8	0.0051	0.0000	0.0007	0.0660	0.6977	0.0190	0.0009	0.0024	0.2104	0.6621				
		9	-0.0004	0.0002	-0.0001	-2.4181	2.1501	0.0203	0.0008	0.0026						
445	4	7	0.800	0.0000	-0.000	0.0698	-0.03	0.6803	3.03	0.06	3.09	0.2401	0.6368			
		8	0.0054	0.0000	0.0007	0.0698	0.6803	0.0249	0.0011	0.0031	0.2013	0.6451				
		9	-0.0003	0.0002	-0.0001	-2.6025	2.4653	0.0255	0.0010	0.0032						
445	5	7	0.799	1.03	-0.000	0.1061	-0.03	0.7080	3.04	0.06	3.10	0.2407	0.6358			
		8	0.0052	0.0021	0.0007	0.1061	0.7080	0.0301	0.0014	0.0038	0.2026	0.6506				
		9	-0.0006	0.0002	-0.0001	-1.7846	1.2493	0.0301	0.0012	0.0039						
445	6	7	0.799	3.14	-0.000	0.1019	-0.03	0.7091	3.08	0.06	3.13	0.2504	0.6129			
		8	0.0044	0.0000	0.0006	0.1019	0.7091	0.0483	0.0024	0.0059	0.1997	0.6230				
		9	-0.0006	0.0002	-0.0001	-1.7533	1.5057	0.0494	0.0019	0.0061						
445	7	7	0.801	6.21	-0.000	0.0336	-0.03	0.6931	3.11	0.06	3.17	0.2312	0.5727			
		8	0.0035	0.0000	0.0004	0.0336	0.6931	0.0677	0.0031	0.0077	0.1911	0.5749				
		9	0.0001	0.0002	-0.0000	7.6116	-1.1803	0.0717	0.0027	0.0092						
445	8	7	0.801	9.33	-0.000	0.0339	-0.03	0.7005	3.13	0.06	3.17	0.1743	0.5541			
		8	0.0030	0.0000	0.0004	0.0339	0.7005	0.0816	0.0028	0.0090	0.1374	0.5534				
		9	0.0001	0.0002	-0.0000	10.1890	-2.1091	0.0854	0.0023	0.0094						
445	9	7	0.800	12.48	-0.000	0.0115	-0.03	0.7036	3.12	0.06	3.17	0.1237	0.5601			
		8	0.0033	0.0000	0.0004	0.0115	0.7036	0.0899	0.0022	0.0100	0.1031	0.5645				
		9	-0.0006	0.0002	-0.0001	-1.7830	1.1470	0.0924	0.0019	0.0102						
445	10	7	0.800	-0.02	-0.000	0.0551	-0.02	0.6715	3.02	0.06	3.09	0.2368	0.6470			
		8	0.0053	0.0000	0.0007	0.0551	0.6715	0.0159	0.0068	0.0024	0.2059	0.6494				
		9	-0.0005	0.0002	-0.0001	-2.0426	1.5081	0.0201	0.0008	0.0026						
446	3	7	0.801	-3.04	-0.000	0.0064	-0.02	0.7145	6.02	0.05	6.02	0.2861	0.7618			
		8	0.0036	0.0000	0.0005	0.0064	0.7145	0.0079	0.0004	0.0012	0.1917	0.7094				
		9	-0.0009	0.0001	-0.0002	-0.6349	1.1461	0.0105	0.0004	0.0014						
446	4	7	0.800	-1.02	-0.000	0.0661	-0.01	0.7456	6.06	0.05	6.05	0.2488	0.6551			
		8	0.0046	0.0000	0.0006	0.0661	0.7456	0.0283	0.0014	0.0037	0.1991	0.6552				
		9	-0.0009	0.0001	-0.0001	-0.9663	0.7249	0.0303	0.0012	0.0039						
446	5	7	0.800	-0.00	-0.000	0.0783	-0.01	0.7084	6.08	0.05	6.07	0.2482	0.6450			
		8	0.0051	0.0000	0.0007	0.0783	0.7084	0.0382	0.0018	0.0049	0.1972	0.6497				
		9	-0.0007	0.0002	-0.0001	-1.3450	0.6807	0.0401	0.0015	0.0052						

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																				
446	6	7	0.800	0.58	-0.00	0.0925	-0.01	6.04	0.05	6.08	0.2514	0.6356	0.0053	0.0000	0.0007	0.0007	0.0925	0.7103	0.425	0.021	0.054	0.2514	0.6356
		8	-0.0007	0.0002	0.0001	-1.4294	0.7399	0.0447	0.0017	0.0057	0.1976	0.6426	-0.0007	0.0002	-0.0001	0.0007	-1.4294	0.7399	0.0447	0.0017	0.0057	0.1976	0.6426
446	7	7	0.801	1.06	-0.00	0.1058	-0.01	6.10	0.06	6.08	0.2542	0.6228	0.0052	0.0001	0.0007	0.1058	0.7006	0.467	0.023	0.058	0.2542	0.6228	
		8	-0.0006	0.0002	-0.0001	-1.5696	0.8986	0.0487	0.0019	0.0061	0.2009	0.6307	-0.0006	0.0002	-0.0001	0.0007	-1.5696	0.8986	0.0487	0.0019	0.0061	0.2009	0.6307
446	8	7	0.800	3.13	-0.00	0.1019	-0.01	6.14	0.06	6.12	0.2538	0.5789	0.0044	0.0000	0.0006	0.1019	0.7091	0.605	0.030	0.070	0.2538	0.5789	
		8	-0.0007	0.0002	-0.0001	-1.5864	0.7540	0.0630	0.0025	0.0073	0.2061	0.5869	-0.0007	0.0002	-0.0001	0.0006	-1.5864	0.7540	0.0630	0.0025	0.0073	0.2061	0.5869
446	9	7	0.799	6.23	-0.00	0.0406	-0.02	6.16	0.06	6.13	0.2094	0.5607	0.0035	0.0000	0.0005	0.0406	0.7184	0.792	0.033	0.088	0.2094	0.5607	
		8	-0.0000	0.0002	-0.0000	-19.4232	1.2288	0.0821	0.0027	0.0091	0.1672	0.5555	-0.0000	0.0002	-0.0000	0.0005	-19.4232	1.2288	0.0821	0.0027	0.0091	0.1672	0.5555
446	10	7	0.800	9.31	-0.00	0.0200	-0.02	6.15	0.06	6.12	0.1390	0.5577	0.0033	0.0000	0.0004	0.0200	0.7047	0.855	0.023	0.095	0.1390	0.5577	
		8	-0.0000	0.0002	0.0000	41.4323	3.2619	0.0892	0.0018	0.0098	0.1053	0.5518	-0.0000	0.0002	0.0000	41.4323	3.2619	0.0892	0.0018	0.0098	0.1053	0.5518	
446	11	7	0.801	12.49	-0.00	0.0360	-0.01	6.16	0.06	6.13	0.1184	0.5554	0.0033	0.0000	0.0004	0.0360	0.7442	0.934	0.022	0.103	0.1184	0.5554	
		8	-0.0004	0.0002	-0.0000	-2.3168	0.6959	0.0961	0.0018	0.0106	0.0936	0.5521	-0.0004	0.0002	-0.0000	-2.3168	0.6959	0.0961	0.0018	0.0106	0.0936	0.5521	
446	12	7	0.799	0.00	-0.00	0.0582	-0.01	6.09	0.05	6.08	0.2440	0.6404	0.0053	0.0000	0.0007	0.0582	0.6647	0.382	0.018	0.049	0.2440	0.6404	
		8	-0.0007	0.0002	-0.0001	-1.5483	0.8830	0.0402	0.0015	0.0051	0.1961	0.6410	-0.0007	0.0002	-0.0001	-1.5483	0.8830	0.0402	0.0015	0.0051	0.1961	0.6410	
447	3	7	0.798	-3.05	-0.00	0.0030	-0.02	9.09	0.06	9.02	0.2423	0.6656	0.0037	-0.0001	0.0002	0.0030	0.9324	0.263	0.011	0.037	0.2423	0.6656	
		8	-0.0011	0.0001	-0.0002	-0.7461	0.9324	0.0286	0.0011	0.0037	0.1999	0.6626	-0.0011	0.0001	-0.0002	-0.7461	0.9324	0.0286	0.0011	0.0037	0.1999	0.6626	
447	4	7	0.799	-1.01	-0.00	0.0603	-0.01	9.14	0.06	9.05	0.2494	0.6324	0.0047	0.0000	0.0006	0.0603	0.7260	0.451	0.022	0.057	0.2494	0.6324	
		8	-0.0009	0.0002	-0.0001	-1.1582	0.6842	0.0485	0.0019	0.0061	0.1992	0.6312	-0.0009	0.0002	-0.0001	-1.1582	0.6842	0.0485	0.0019	0.0061	0.1992	0.6312	
447	5	7	0.799	0.03	-0.00	0.0677	-0.01	9.16	0.05	9.07	0.2603	0.6081	0.0053	0.0000	0.0007	0.0677	0.6747	0.528	0.027	0.064	0.2603	0.6081	
		8	-0.0008	0.0002	-0.0001	-1.3264	0.7719	0.0554	0.0022	0.0067	0.2063	0.6063	-0.0008	0.0002	-0.0001	-1.3264	0.7719	0.0554	0.0022	0.0067	0.2063	0.6063	
447	6	7	0.799	0.59	-0.00	0.0774	-0.01	9.16	0.05	9.07	0.2615	0.5951	0.0053	0.0000	0.0007	0.0774	0.6848	0.559	0.029	0.066	0.2615	0.5951	
		8	-0.0006	0.0002	-0.0001	-1.6819	0.8582	0.0591	0.0024	0.0070	0.2052	0.5961	-0.0006	0.0002	-0.0001	-1.6819	0.8582	0.0591	0.0024	0.0070	0.2052	0.5961	

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd																			
447	7	0.800	1.07	-0.00	0.0953	-0.01	9.16	0.05	9.08	0.2537	0.5871									
	8	0.0053	0.0001	0.0007	0.0953	0.6756	0.0589	0.0029	0.0069	0.2028	0.5871									
	9	-0.0007	0.0002	-0.0001	-1.4736	0.7554	0.0621	0.0025	0.0072											
447	8	0.800	3.16	-0.00	0.0996	-0.01	9.19	0.06	9.09	0.2184	0.5735									
	8	0.0044	0.0000	0.0006	0.0996	0.7176	0.0746	0.0032	0.0085	0.1778	0.5735									
	9	-0.0010	0.0002	-0.0001	-1.1644	0.4739	0.0765	0.0027	0.0087											
447	9	0.799	6.24	-0.00	0.0431	-0.01	9.19	0.06	9.09	0.1628	0.5588									
	9	0.0034	0.0000	0.0004	0.0431	0.7216	0.0835	0.0027	0.0093	0.1289	0.5588									
	9	-0.0000	0.0002	-0.0000	-18.2499	0.5579	0.0867	0.0022	0.0095											
447	10	0.798	9.33	-0.00	0.0163	-0.01	9.18	0.06	9.08	0.1202	0.5632									
	8	0.0033	0.0000	0.0004	0.0163	0.7159	0.0891	0.0021	0.0100	0.0941	0.5632									
	9	-0.0002	0.0002	0.0000	-5.1999	-0.1712	0.0925	0.0017	0.0102											
447	11	0.798	12.49	-0.00	0.0210	-0.01	9.19	0.06	9.08	0.1059	0.5619									
	8	0.0033	0.0000	0.0004	0.0210	0.7280	0.0975	0.0020	0.0109	0.0854	0.5619									
	9	-0.0007	0.0002	-0.0000	-1.4681	0.5554	0.0998	0.0017	0.0110											
447	12	0.800	0.02	-0.00	0.0636	-0.01	9.16	0.06	9.07	0.2586	0.6076									
	8	0.0052	0.0000	0.0007	0.0636	0.6885	0.0528	0.0027	0.0064	0.2059	0.6076									
	9	-0.0008	0.0002	-0.0001	-1.4524	0.6040	0.0554	0.0022	0.0066											
448	1	0.798	3.02	-0.00	0.0029	-0.01	12.07	0.06	12.08	0.2449	0.6431									
	8	0.0036	0.0000	0.0005	0.0029	0.7248	0.0433	0.0021	0.0055	0.1990	0.6431									
	9	-0.0013	0.0001	-0.0001	-0.7316	0.6938	0.0476	0.0018	0.0060											
448	2	0.799	-0.99	-0.00	0.0454	-0.01	12.10	0.05	12.10	0.2492	0.5937									
	8	0.0045	0.0000	0.0006	0.0454	0.7076	0.0565	0.0028	0.0067	0.1971	0.5937									
	9	-0.0010	0.0002	-0.0001	-1.0464	0.7720	0.0609	0.0024	0.0071											
448	3	0.799	-0.98	-0.00	0.0572	-0.01	12.11	0.06	12.10	0.2526	0.5904									
	8	0.0047	0.0000	0.0006	0.0572	0.7174	0.0575	0.0029	0.0068	0.2014	0.5904									
	9	-0.0008	0.0002	-0.0001	-1.3035	0.8847	0.0619	0.0024	0.0073											
448	4	0.799	0.04	-0.00	0.0660	-0.01	12.12	0.06	12.12	0.2309	0.5849									
	8	0.0051	0.0000	0.0007	0.0660	0.6977	0.0652	0.0030	0.0076	0.1861	0.5849									
	9	-0.0008	0.0002	-0.0001	-1.3939	0.6744	0.0689	0.0025	0.0080											
448	5	0.800	0.59	-0.00	0.0854	-0.01	12.13	0.06	12.11	0.2216	0.5804									
	8	0.0052	0.0000	0.0007	0.0854	0.7094	0.0696	0.0030	0.0081	0.1774	0.5804									
	9	-0.0007	0.0002	-0.0001	-1.5327	0.7078	0.0727	0.0025	0.0084											

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key														
448	6	0.800	1.10	-0.00	0.0767	-0.01	12.13	0.06	12.11	0.2149	0.5792					
	6	0.0053	0.0000	0.0007	0.0000	0.6702	0.0735	0.0031	0.0085	0.1725	0.5761					
	9	-0.0007	0.0002	-0.0000	-1.6108	0.5941	0.0760	0.0026	0.0087							
448	7	0.800	5.17	-0.00	0.0985	-0.01	12.13	0.06	12.12	0.1885	0.5568					
	6	0.0044	0.0000	0.0006	0.0000	0.6910	0.0805	0.0030	0.0089	0.1524	0.5548					
	9	-0.0009	0.0002	-0.0000	-1.4220	0.5468	0.0838	0.0025	0.0093							
448	8	0.798	6.25	-0.00	0.0265	-0.01	12.12	0.06	12.12	0.1311	0.5601					
	6	0.0035	0.0000	0.0004	0.0000	0.7030	0.0862	0.0022	0.0096	0.1008	0.5556					
	9	-0.0001	0.0002	-0.0000	-9.0440	0.0101	0.0895	0.0018	0.0099							
448	9	0.798	9.34	-0.00	0.0200	-0.01	12.13	0.06	12.11	0.1182	0.5592					
	6	0.0033	0.0000	0.0004	0.0000	0.7047	0.0938	0.0022	0.0104	0.0884	0.5566					
	9	-0.0002	0.0002	-0.0000	-5.7474	0.1816	0.0973	0.0017	0.0108							
448	10	0.798	12.50	-0.00	-0.0049	-0.01	12.14	0.06	12.11	0.0967	0.5627					
	6	0.0035	0.0000	0.0004	0.0000	0.7051	0.1009	0.0019	0.0113	0.0756	0.5554					
	9	-0.0008	0.0002	-0.0000	-1.4961	0.4180	0.1039	0.0015	0.0115							
448	11	0.798	0.02	-0.00	0.0653	-0.00	12.12	0.06	12.11	0.2291	0.5824					
	6	0.0053	0.0000	0.0007	0.0000	0.6742	0.0654	0.0030	0.0076	0.1864	0.5857					
	9	-0.0007	0.0002	-0.0001	-1.5560	0.7843	0.0688	0.0025	0.0080							
449	1	0.798	-2.99	-0.00	0.0155	-0.01	15.11	0.05	15.10	0.2473	0.5928					
	6	0.0037	0.0000	0.0005	0.0000	0.7467	0.0569	0.0028	0.0067	0.2006	0.5945					
	9	-0.0014	0.0002	-0.0001	-0.7683	0.6630	0.0614	0.0024	0.0073							
449	2	0.799	-0.96	-0.00	0.0555	-0.00	15.13	0.05	15.12	0.2125	0.5809					
	6	0.0046	0.0000	0.0006	0.0000	0.7423	0.0728	0.0030	0.0084	0.1694	0.5811					
	9	-0.0007	0.0002	-0.0001	-1.5220	0.9720	0.0761	0.0025	0.0088							
449	3	0.799	0.06	-0.00	0.0583	-0.01	15.14	0.06	15.12	0.2048	0.5736					
	6	0.0052	0.0000	0.0007	0.0000	0.6802	0.0793	0.0032	0.0090	0.1604	0.5713					
	9	-0.0007	0.0002	-0.0001	-1.5993	0.8649	0.0816	0.0026	0.0093							
449	4	0.800	0.62	-0.00	0.0824	-0.01	15.14	0.06	15.12	0.1979	0.5634					
	6	0.0052	0.0000	0.0007	0.0000	0.6942	0.0812	0.0032	0.0091	0.1567	0.5634					
	9	-0.0007	0.0002	-0.0001	-1.6695	0.7851	0.0835	0.0026	0.0094							
449	5	0.800	1.10	-0.00	0.0813	-0.00	15.14	0.06	15.12	0.1923	0.5583					
	6	0.0052	0.0000	0.0007	0.0000	0.6726	0.0815	0.0031	0.0091	0.1526	0.5596					
	9	-0.0008	0.0002	-0.0001	-1.5413	0.6236	0.0839	0.0025	0.0093							

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key														
449	6	0.798	5.17	-0.00	0.1047	-0.01	15.13	0.06	15.11	0.1473	0.5576					
	6	0.0044	0.0000	0.0006	0.1047	0.7024	0.0854	0.0025	0.0095	0.1190	0.5553					
	9	-0.0008	0.0002	-0.0001	-1.4429	0.8754	0.0886	0.0021	0.0098							
449	7	0.797	6.25	-0.00	0.0283	-0.01	15.13	0.06	15.11	0.1195	0.5620					
	6	0.0034	0.0000	0.0004	0.0283	0.7058	0.0908	0.0021	0.0102	0.0936	0.5577					
	9	-0.0003	0.0002	-0.0000	-4.4893	0.4200	0.0942	0.0017	0.0105							
449	8	0.800	9.35	-0.00	0.0210	-0.01	15.14	0.06	15.12	0.1052	0.5622					
	6	0.0033	0.0000	0.0004	0.0210	0.7280	0.0986	0.0020	0.0110	0.0823	0.5563					
	9	-0.0001	0.0002	0.0000	-7.4717	-0.0365	0.1013	0.0016	0.0112							
449	9	0.798	12.51	-0.00	-0.0058	-0.01	15.13	0.06	15.12	0.0896	0.5630					
	6	0.0037	-0.0000	0.0005	-0.0058	0.7316	0.1056	0.0018	0.0116	0.0693	0.5570					
	9	-0.0008	0.0002	-0.0000	-1.3926	0.3002	0.1086	0.0015	0.0120							
449	10	0.798	0.05	-0.00	0.0681	-0.01	15.14	0.06	15.12	0.2027	0.5708					
	6	0.0052	0.0000	0.0007	0.0681	0.6905	0.0793	0.0032	0.0090	0.1602	0.5702					
	9	-0.0009	0.0002	-0.0001	-1.4098	0.6021	0.0815	0.0026	0.0093							
450	3	0.601	-3.13	-0.00	0.0625	-0.02	-0.11	0.05	-0.06	0.2642	0.6202					
	6	0.0031	0.0000	0.0004	0.0625	0.7732	-0.0234	-0.0012	-0.0029	0.3039	0.6201					
	9	-0.0012	0.0000	-0.0002	-0.2037	0.9464	-0.0192	-0.0011	-0.0024							
450	4	0.601	-1.07	-0.00	0.1026	-0.01	-0.07	0.05	-0.04	0.2152	0.7061					
	6	0.0046	0.0000	0.0006	0.1026	0.7038	-0.0051	-0.0002	-0.0007	0.5331	0.6573					
	9	-0.0005	0.0000	-0.0002	-0.3356	1.9614	-0.0031	-0.0003	-0.0004							
450	5	0.601	-0.03	-0.00	0.1107	-0.01	-0.06	0.05	-0.02	0.6649	0.6625					
	6	0.0053	0.0001	0.0007	0.1107	0.6590	0.0015	0.0002	0.0002	0.0449	0.7679					
	9	-0.0004	0.0000	-0.0001	-0.4792	1.9958	0.0025	0.0000	0.0003							
450	6	0.600	0.52	-0.00	0.1240	-0.01	-0.05	0.05	-0.01	0.4136	0.7036					
	6	0.0052	0.0001	0.0006	0.1240	0.6449	0.0051	0.0004	0.0007	0.1679	0.7316					
	9	-0.0005	0.0000	-0.0001	-0.4186	1.8230	0.0058	0.0001	0.0008							
450	7	0.600	1.00	-0.00	0.1381	-0.01	-0.04	0.05	-0.01	0.3494	0.6984					
	6	0.0052	0.0001	0.0007	0.1381	0.6832	0.0089	0.0006	0.0012	0.2052	0.6930					
	9	-0.0004	0.0000	-0.0001	-0.4905	2.0567	0.0090	0.0003	0.0012							
450	8	0.600	3.08	-0.00	0.1205	-0.01	-0.00	0.05	-0.01	0.2971	0.6345					
	6	0.0046	0.0001	0.0005	0.1205	0.6447	0.0274	0.0016	0.0034	0.2268	0.6470					
	9	-0.0003	0.0000	-0.0001	-0.6517	2.4878	0.0265	0.0012	0.0034							

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd																			
450	9	7	0.599	6.16	-0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00
		6	0.0031	0.0000	0.0004	0.0031	0.7510	0.0514	0.0029	0.0061	0.2061	0.5964	0.2332	0.0066	0.0029	0.0061	0.0066	0.0066	0.0066	0.0066	0.0066
		9	0.0004	0.0001	-0.0000	1.2576	-0.4855	0.0554	0.0025	0.0066	0.2332	0.6044	0.2332	0.0066	0.0025	0.0066	0.0066	0.0066	0.0066	0.0066	0.0066
450	10	7	0.600	9.25	-0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00
		6	0.0029	0.0000	0.0004	0.0792	0.7705	0.0720	0.0038	0.0081	0.2706	0.5643	0.2227	0.0081	0.0038	0.0081	0.0081	0.0081	0.0081	0.0081	0.0081
		9	0.0003	0.0000	-0.0000	1.0135	-0.6804	0.0772	0.0034	0.0087	0.2227	0.5636	0.2227	0.0087	0.0034	0.0087	0.0087	0.0087	0.0087	0.0087	0.0087
450	11	7	0.600	12.41	-0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00
		6	0.0030	0.0000	0.0004	0.0762	0.7930	0.0846	0.0034	0.0092	0.2017	0.5472	0.1709	0.0092	0.0034	0.0092	0.0092	0.0092	0.0092	0.0092	0.0092
		9	-0.0000	0.0000	-0.0001	-3.2002	9.4859	0.0892	0.0030	0.0096	0.1709	0.5421	0.1709	0.0096	0.0030	0.0096	0.0096	0.0096	0.0096	0.0096	0.0096
450	12	7	0.599	-0.06	-0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00
		6	0.0050	0.0001	0.0006	0.1110	0.6849	0.0015	0.0001	0.0001	0.5706	0.4824	0.0145	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001
		9	-0.0007	0.0000	-0.0001	-0.4782	1.2384	0.0025	0.0000	0.0003	0.0145	0.7002	0.0145	0.0003	0.0000	0.0003	0.0003	0.0003	0.0003	0.0003	0.0003
451	1	7	0.598	-3.10	-0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00
		6	0.0031	0.0000	0.0004	0.0625	0.7732	-0.0405	0.0021	-0.0050	0.2591	0.6250	0.2646	-0.0050	0.0021	-0.0050	-0.0050	-0.0050	-0.0050	-0.0050	-0.0050
		9	-0.0013	0.0000	-0.0002	-0.1950	0.9385	-0.0371	-0.0019	-0.0047	0.2646	0.6379	0.2646	-0.0047	-0.0019	-0.0047	-0.0047	-0.0047	-0.0047	-0.0047	-0.0047
451	2	7	0.598	-1.09	-0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00
		6	0.0045	0.0000	0.0006	0.1032	0.7056	-0.0209	0.0011	-0.0027	0.2628	0.6464	0.3033	-0.0027	0.0011	-0.0027	-0.0027	-0.0027	-0.0027	-0.0027	-0.0027
		9	-0.0007	0.0000	-0.0002	-0.4177	1.3513	-0.0183	-0.0011	-0.0023	0.3033	0.6449	0.3033	-0.0023	-0.0011	-0.0023	-0.0023	-0.0023	-0.0023	-0.0023	-0.0023
451	3	7	0.599	-0.09	-0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00
		6	0.0050	0.0001	0.0006	0.1106	0.6763	-0.0118	0.0005	-0.0015	0.2428	0.6755	0.3510	-0.0015	0.0005	-0.0015	-0.0015	-0.0015	-0.0015	-0.0015	-0.0015
		9	-0.0004	0.0000	-0.0001	-0.6815	2.1741	-0.0100	-0.0007	-0.0013	0.3510	0.6713	0.3510	-0.0013	-0.0007	-0.0013	-0.0013	-0.0013	-0.0013	-0.0013	-0.0013
451	4	7	0.601	0.50	-0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00
		6	0.0050	0.0001	0.0006	0.1329	0.6913	-0.0073	0.0003	-0.0010	0.2343	0.7244	0.3833	-0.0010	0.0003	-0.0010	-0.0010	-0.0010	-0.0010	-0.0010	-0.0010
		9	-0.0006	0.0000	-0.0001	-0.5963	1.2199	-0.0065	-0.0003	-0.0008	0.3833	0.6357	0.3833	-0.0008	-0.0003	-0.0008	-0.0008	-0.0008	-0.0008	-0.0008	-0.0008
451	5	7	0.600	0.97	-0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00
		6	0.0053	0.0001	0.0006	0.1155	0.6445	-0.0037	0.0001	-0.0005	0.1926	0.7940	0.5343	-0.0005	0.0001	-0.0005	-0.0005	-0.0005	-0.0005	-0.0005	-0.0005
		9	-0.0004	0.0000	-0.0001	-0.6978	1.8186	-0.0031	-0.0003	-0.0004	0.5343	0.7723	0.5343	-0.0004	-0.0003	-0.0004	-0.0004	-0.0004	-0.0004	-0.0004	-0.0004
451	6	7	0.600	3.05	-0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00
		6	0.0044	0.0001	0.0005	0.1254	0.6648	0.0103	0.0007	-0.0013	0.3462	0.6516	0.2121	-0.0013	0.0007	-0.0013	-0.0013	-0.0013	-0.0013	-0.0013	-0.0013
		9	-0.0003	0.0000	-0.0001	-0.7904	2.4312	0.0099	0.0004	0.0013	0.2121	0.6531	0.2121	0.0013	0.0004	0.0013	0.0013	0.0013	0.0013	0.0013	0.0013
451	7	7	0.600	6.17	-0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00	0.00
		6	0.0030	0.0000	0.0004	0.0883	0.7704	-0.0364	0.0020	-0.0045	0.2874	0.6316	0.2874	-0.0045	0.0020	-0.0045	-0.0045	-0.0045	-0.0045	-0.0045	-0.0045
		9	0.0002	0.0001	-0.0000	2.1279	-1.0627	0.0388	0.0018	0.0049	0.2874	0.6335	0.2874	0.0049	0.0018	0.0049	0.0049	0.0049	0.0049	0.0049	0.0049

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	←	→	See Key	→	←	→	←	→	←	→	←	→	←	→	←	→	←	→
451	8	7	0.600	9.25	-0.00	0.0749	-0.02	-2.81	0.06	-2.85	0.2877	0.5821								
	6	9	0.0030	0.0000	0.0004	1.4087	0.7848	0.0522	0.0034	0.0069	0.2361	0.5879								
	9		0.0003	0.0001	-0.0000		-0.1743	0.0639	0.0030	0.0075										
451	9	7	0.598	12.39	-0.00	0.0656	-0.02	-2.79	0.05	-2.83	0.2500	0.5524								
	8	9	0.0030	0.0000	0.0004	7.5686	0.7847	0.0777	0.0038	0.0085	0.2104	0.5540								
	9		0.0000	0.0000	-0.0001		-26.5742	0.0830	0.0034	0.0092										
451	10	7	0.599	-0.07	-0.00	0.1069	-0.01	-2.96	0.05	-2.98	0.2436	0.6772								
	8	9	0.0051	0.0001	0.0006	-0.6442	0.6665	-0.0119	-0.0005	-0.0016	0.3455	0.6721								
	9		-0.0008	0.0001	-0.0001		-0.8625	-0.0102	-0.0007	-0.0013										
452	1	7	0.599	-3.09	-0.00	0.0627	-0.02	-1.01	0.05	-1.05	0.2594	0.6274								
	8	9	0.0032	0.0000	0.0004	-0.2356	0.7543	-0.0285	-0.0014	-0.0032	0.2896	0.6427								
	9		-0.0012	0.0000	-0.0002		-0.8750	-0.0250	-0.0014	-0.0032										
452	2	7	0.599	-1.08	-0.00	0.1127	-0.01	-0.97	0.05	-1.01	0.2302	0.6585								
	8	9	0.0044	0.0001	0.0006	-0.2680	0.7089	-0.0098	-0.0004	-0.0012	0.3682	0.6363								
	9		-0.0006	0.0000	-0.0002		-1.7145	-0.0079	-0.0005	-0.0010										
452	3	7	0.600	-0.06	-0.00	0.1075	-0.01	-0.95	0.05	-1.00	0.0090	0.7146								
	8	9	0.0053	0.0001	0.0006	-0.3889	0.6422	-0.0022	-0.0000	-0.0002	0.9035	0.9095								
	9		-0.0006	0.0000	-0.0001		-1.1703	-0.0013	-0.0002	-0.0002										
452	4	7	0.600	0.49	-0.00	0.1353	-0.01	-0.95	0.05	-1.00	0.8481	0.5592								
	8	9	0.0050	0.0001	0.0007	-0.5126	0.6918	0.0011	0.0001	0.0001	0.1535	0.6989								
	9		-0.0005	0.0000	-0.0001		-1.7095	0.0013	-0.0000	0.0001										
452	5	7	0.600	0.97	-0.00	0.1372	-0.01	-0.94	0.05	-1.00	0.4666	0.7153								
	8	9	0.0052	0.0001	0.0007	-0.5447	0.6668	0.0043	0.0004	0.0006	0.1249	0.7091								
	9		-0.0005	0.0000	-0.0001		-1.5028	0.0043	0.0001	0.0006										
452	6	7	0.601	3.07	-0.00	0.1277	-0.01	-0.91	0.05	-0.97	0.3112	0.6353								
	8	9	0.0045	0.0001	0.0005	-0.6157	0.6560	0.0218	0.0013	0.0026	0.2339	0.6550								
	9		-0.0005	0.0000	-0.0001		-1.6638	0.0204	0.0009	0.0026										
452	7	7	0.601	6.16	-0.00	0.0824	-0.02	-0.85	0.06	-0.92	0.2907	0.6132								
	8	9	0.0031	0.0000	0.0004	1.0386	0.7372	0.0465	0.0027	0.0057	0.2344	0.6211								
	9		0.0004	0.0001	-0.0000		-0.3329	0.0498	0.0023	0.0061										
452	8	7	0.599	9.27	-0.00	0.0650	-0.02	-0.82	0.05	-0.88	0.2766	0.5734								
	8	9	0.0030	0.0000	0.0004	1.6155	0.7503	0.0686	0.0037	0.0078	0.2280	0.5778								
	9		0.0002	0.0000	-0.0000		-0.3454	0.0737	0.0033	0.0085										

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	y	Cd																		
452	7	0.600	12.40	-0.000	0.0226	-0.002	-0.01	0.05	-0.088	0.2185	0.5478									
	6	0.0028	0.0000	0.0004	0.7737	0.0827	0.0036	0.0090	0.0090	0.1857	0.5450									
	5	-0.0007	0.0000	-0.0001	-0.2577	1.1037	0.0871	0.0032	0.0095	0.1857	0.5450									
452	10	0.599	-0.04	-0.000	0.1224	-0.001	-0.096	0.06	-1.00	0.0977	0.9262									
	8	0.0050	0.0001	0.0007	0.6967	-0.0018	-0.0003	0.0000	-0.0003	0.0977	0.9262									
	5	-0.0007	0.0000	-0.0001	-0.5633	1.0088	-0.0010	-0.0002	-0.0002	1.0206	1.0359									
453	1	0.599	-3.08	-0.000	-0.02	0.96	0.05	0.05	0.92	0.2621	0.6300									
	6	0.0035	0.0000	0.0004	0.0527	0.6708	-0.0168	-0.0008	-0.0021	0.3227	0.6095									
	5	-0.0009	0.0000	-0.0002	-0.2466	1.1591	-0.0139	-0.0008	-0.0016	0.3227	0.6095									
453	2	0.601	-1.07	-0.000	-0.01	1.00	0.05	0.05	0.95	-0.6313	0.8080									
	6	0.0047	0.0000	0.0006	0.0956	0.6828	-0.0005	0.0000	-0.0000	-0.6313	0.8080									
	9	-0.0007	0.0000	-0.0002	-0.1914	1.4343	0.0006	-0.0001	0.0000	-0.6313	0.8080									
453	3	0.600	-0.05	-0.000	-0.01	1.01	0.05	0.05	0.96	0.4501	0.7759									
	6	0.0050	0.0001	0.0007	0.1248	0.6971	0.0056	0.0005	0.0008	0.4501	0.7759									
	9	-0.0004	0.0000	-0.0001	-0.4669	1.9296	0.0064	0.0002	0.0009	0.4501	0.7759									
453	4	0.600	0.44	-0.000	-0.01	1.02	0.05	0.05	0.96	0.3521	0.7153									
	6	0.0050	0.0001	0.0007	0.1358	0.7004	0.0100	0.0007	0.0014	0.3521	0.7153									
	9	-0.0004	0.0000	-0.0001	-0.5548	2.2113	0.0101	0.0004	0.0014	0.3521	0.7153									
453	5	0.600	1.00	-0.000	-0.01	1.03	0.05	0.05	0.97	0.3198	0.6701									
	6	0.0052	0.0001	0.0006	0.1244	0.6631	0.0146	0.0009	0.0019	0.3198	0.6701									
	9	-0.0005	0.0000	-0.0001	-0.4639	1.6411	0.0141	0.0006	0.0018	0.3198	0.6701									
453	6	0.600	3.09	-0.000	-0.01	1.07	0.05	0.05	1.00	0.2920	0.5364									
	6	0.0046	0.0001	0.0006	0.1361	0.6676	0.0339	0.0019	0.0043	0.2920	0.5364									
	9	-0.0004	0.0000	-0.0001	-0.9221	1.9121	0.0325	0.0014	0.0042	0.2920	0.5364									
453	7	0.600	6.17	-0.000	-0.02	1.11	0.05	0.05	1.05	0.2384	0.5931									
	6	0.0032	0.0000	0.0004	0.0727	0.7058	0.0557	0.0032	0.0066	0.2384	0.5931									
	9	0.0004	0.0001	-0.0000	-1.3466	-0.4848	0.0592	0.0028	0.0071	0.2384	0.5931									
453	8	0.600	9.27	-0.000	-0.01	1.15	0.05	0.05	1.07	0.2647	0.5588									
	6	0.0029	0.0000	0.0004	0.0785	0.7559	0.0746	0.0039	0.0083	0.2647	0.5588									
	9	0.0001	0.0001	-0.0000	-1.7465	-1.5816	0.0794	0.0034	0.0088	0.2647	0.5588									
453	9	0.600	12.42	-0.000	-0.02	1.14	0.05	0.05	1.07	0.1867	0.5507									
	6	0.0029	0.0000	0.0004	0.0704	0.7855	0.0858	0.0032	0.0094	0.1867	0.5507									
	9	-0.0004	0.0000	-0.0001	-0.4919	1.5603	0.0903	0.0028	0.0098	0.1867	0.5507									

See Key

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																			
453	10	7	0.598	-0.002	-0.007	0.1224	-0.01	1.02	0.05	0.96	0.4213	0.7263	0.0050	0.0001	0.0007	0.1224	0.6967	0.0058	0.0004	0.0008	0.1638	0.6890
		8	-0.0050	0.0001	0.0007	0.4964	0.6967	0.0058	0.0002	0.0008												
		9	-0.0006	0.0000	-0.0001	-0.4964	1.4631	0.0068	0.0002	0.0009												
454	1	7	0.600	-3.07	0.0004	0.0534	-0.01	2.96	0.05	3.04	0.2137	0.6355	0.0036	0.0000	0.0004	0.0534	0.6579	-0.0066	-0.0008	0.5055	0.6092	
		8	-0.0036	0.0000	0.0004	-0.1988	0.6579	-0.0066	-0.0003	-0.0004												
		9	-0.0009	0.0000	-0.0002	-0.1988	1.2249	-0.0034	-0.0003	-0.0004												
454	2	7	0.601	-1.06	0.0006	0.1037	-0.01	2.99	0.05	3.06	0.3748	0.7673	0.0044	0.0000	0.0006	0.1037	0.7153	0.0076	0.0005	0.2030	0.6933	
		8	-0.0044	0.0000	0.0006	-0.2697	0.7153	0.0076	0.0004	0.0013												
		9	-0.0008	0.0000	-0.0001	-0.2697	1.1609	0.0098	0.0004	0.0013												
454	3	7	0.601	-0.02	0.0007	0.1083	-0.01	3.00	0.05	3.07	0.3118	0.6715	0.0053	0.0001	0.0007	0.1083	0.6584	0.0166	0.0010	0.2229	0.6533	
		8	-0.0053	0.0001	0.0007	-0.4639	0.6584	0.0176	0.0007	0.0023												
		9	-0.0005	0.0000	-0.0001	-0.4639	1.6411	0.0219	0.0009	0.0028												
454	4	7	0.601	0.53	0.0007	0.1248	-0.01	3.02	0.05	3.08	0.2979	0.6495	0.0052	0.0001	0.0007	0.1248	0.6714	0.0213	0.0012	0.2223	0.6495	
		8	-0.0052	0.0001	0.0007	-0.4923	0.6714	0.0213	0.0012	0.0028												
		9	-0.0006	0.0000	-0.0001	-0.4923	1.4311	0.0219	0.0009	0.0028												
454	5	7	0.599	1.00	0.0007	0.1334	-0.01	3.03	0.05	3.09	0.2744	0.5837	0.0050	0.0001	0.0007	0.1334	0.6999	0.0263	0.0015	0.2223	0.6495	
		8	-0.0050	0.0001	0.0007	-0.4880	0.6999	0.0263	0.0015	0.0034												
		9	-0.0006	0.0000	-0.0001	-0.4880	1.3981	0.0264	0.0011	0.0034												
454	6	7	0.601	5.09	0.0006	0.1268	-0.01	3.07	0.06	3.12	0.2901	0.6215	0.0046	0.0001	0.0006	0.1268	0.6649	0.0445	0.0025	0.2296	0.6322	
		8	-0.0046	0.0001	0.0006	-0.6677	0.6649	0.0445	0.0025	0.0056												
		9	-0.0006	0.0000	-0.0001	-0.6677	1.0534	0.0450	0.0020	0.0056												
454	7	7	0.600	6.17	0.0004	0.0818	-0.02	3.11	0.05	3.16	0.2744	0.5837	0.0031	0.0002	0.0004	0.0818	0.7235	0.0648	0.0035	0.2257	0.5837	
		8	-0.0031	0.0000	0.0004	-2.3045	0.7235	0.0701	0.0031	0.0061												
		9	-0.0002	0.0001	-0.0000	-2.3045	-0.4776	0.0701	0.0031	0.0061												
454	8	7	0.600	9.28	0.0004	0.0605	-0.01	3.12	0.06	3.18	0.2334	0.5506	0.0028	0.0001	0.0004	0.0605	0.7859	0.0803	0.0037	0.1956	0.5477	
		8	-0.0028	0.0000	0.0004	-7.3174	0.7859	0.0803	0.0037	0.0063												
		9	-0.0000	0.0001	-0.0000	-7.3174	3.4834	0.0853	0.0033	0.0093												
454	9	7	0.599	12.41	0.0004	0.0177	-0.01	3.11	0.05	3.16	0.1666	0.5474	0.0030	0.0001	0.0004	0.0177	0.849	0.0893	0.0028	0.1392	0.5474	
		8	-0.0030	0.0000	0.0004	-0.3045	0.7095	0.0849	0.0028	0.0093												
		9	-0.0010	0.0000	-0.0001	-0.3045	0.7564	0.0893	0.0024	0.0097												
454	10	7	0.599	-0.02	0.0007	0.1025	-0.01	3.01	0.05	3.08	0.3054	0.6612	0.0054	0.0001	0.0007	0.1025	0.6487	0.0163	0.0010	0.2245	0.6612	
		8	-0.0054	0.0001	0.0007	-0.6796	0.6487	0.0163	0.0010	0.0023												
		9	-0.0004	0.0000	-0.0001	-0.6796	2.1434	0.0173	0.0007	0.0023												

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																	
455	1	7	0.598	-3.06	-0.00	0.004	0.0574	-0.01	6.00	0.06	6.00	0.0010	0.3452	0.7434					
		8	0.0035	0.0000	0.0004	0.0002	0.0004	0.6839	0.0069	0.0004	0.0010	0.0013	0.2140	0.7136					
		9	-0.0011	0.0000	-0.0002	-0.0002	-0.2240	0.9646	0.0094	0.0004	0.0013	0.0013	0.2140	0.7136					
455	2	7	0.599	-1.02	-0.00	0.006	0.1140	-0.01	6.04	0.05	6.03	0.0033	0.2091	0.6540					
		8	0.0046	0.0001	0.0006	0.0001	0.0006	0.6980	0.0253	0.0014	0.0033	0.0035	0.2232	0.6604					
		9	-0.0008	0.0000	-0.0001	-0.0001	-0.4076	1.1453	0.0271	0.0012	0.0035	0.0035	0.2232	0.6604					
455	3	7	0.600	-0.01	-0.00	0.007	0.1272	-0.01	6.07	0.05	6.05	0.0044	0.2048	0.6433					
		8	0.0051	0.0001	0.0007	0.0001	0.0007	0.6975	0.0349	0.0019	0.0044	0.0046	0.2210	0.6512					
		9	-0.0006	0.0000	-0.0001	-0.0001	-0.5240	1.3123	0.0359	0.0015	0.0046	0.0046	0.2210	0.6512					
455	4	7	0.599	0.54	-0.00	0.007	0.1103	-0.01	6.07	0.06	6.06	0.0050	0.2058	0.6375					
		8	0.0054	0.0001	0.0007	0.0001	0.0007	0.6510	0.0393	0.0022	0.0050	0.0052	0.2234	0.6461					
		9	-0.0007	0.0000	-0.0001	-0.0001	-0.4705	1.1767	0.0403	0.0018	0.0052	0.0052	0.2234	0.6461					
455	5	7	0.599	1.04	-0.00	0.007	0.1163	-0.01	6.08	0.05	6.06	0.0054	0.2901	0.6320					
		8	0.0053	0.0001	0.0007	0.0001	0.0007	0.6607	0.0433	0.0025	0.0054	0.0056	0.2237	0.6362					
		9	-0.0005	0.0000	-0.0001	-0.0001	-0.5555	1.6021	0.0443	0.0019	0.0056	0.0056	0.2237	0.6362					
455	6	7	0.601	3.13	-0.00	0.005	0.1232	-0.01	6.11	0.06	6.09	0.0068	0.2019	0.5917					
		8	0.0046	0.0001	0.0005	0.0001	0.0005	0.6455	0.0582	0.0032	0.0068	0.0072	0.2302	0.6023					
		9	-0.0002	0.0000	-0.0001	-0.0001	-1.5947	3.7008	0.0600	0.0027	0.0072	0.0072	0.2302	0.6023					
455	7	7	0.601	6.19	-0.00	0.004	0.0818	-0.02	6.14	0.06	6.12	0.0084	0.2584	0.5599					
		8	0.0031	0.0000	0.0004	0.0004	0.0004	0.7235	0.0755	0.0039	0.0084	0.0089	0.2132	0.5599					
		9	0.0002	0.0001	-0.0000	-0.0000	3.4359	-0.4690	0.0801	0.0034	0.0089	0.0089	0.2132	0.5599					
455	8	7	0.601	9.26	-0.00	0.004	0.0933	-0.01	6.14	0.06	6.11	0.0093	0.1866	0.5508					
		8	0.0030	0.0000	0.0004	0.0004	0.0004	0.7845	0.0852	0.0031	0.0093	0.0098	0.1548	0.5473					
		9	0.0004	0.0000	-0.0000	-0.0000	1.0973	-0.4867	0.0901	0.0027	0.0098	0.0098	0.1548	0.5473					
455	9	7	0.601	12.43	-0.00	0.004	0.0650	-0.02	6.12	0.05	6.10	0.0092	0.1516	0.5476					
		8	0.0030	0.0000	0.0004	0.0004	0.0004	0.7303	0.0845	0.0025	0.0092	0.0095	0.1204	0.5417					
		9	-0.0006	0.0000	-0.0001	-0.0001	-0.5325	1.0184	0.0882	0.0021	0.0095	0.0095	0.1204	0.5417					
455	10	7	0.599	0.00	-0.00	0.007	0.1338	-0.01	6.07	0.06	6.05	0.0044	0.2802	0.6356					
		8	0.0050	0.0001	0.0007	0.0007	0.0007	0.7086	0.0348	0.0019	0.0044	0.0046	0.2204	0.6505					
		9	-0.0006	0.0000	-0.0001	-0.0001	-0.6595	1.3663	0.0357	0.0015	0.0046	0.0046	0.2204	0.6505					
456	3	7	0.599	-2.99	-0.00	0.004	0.0659	-0.02	15.10	0.06	15.09	0.0067	0.2648	0.6054					
		8	0.0034	0.0000	0.0004	0.0004	0.0004	0.7121	0.0553	0.0029	0.0067	0.0073	0.2176	0.6133					
		9	-0.0012	0.0001	-0.0002	-0.0002	-0.4273	0.8317	0.0599	0.0026	0.0073	0.0073	0.2176	0.6133					

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																						
456	4	7	0.599	-0.97	-0.00	0.0962	-0.02	15.12	0.05	15.11	0.2509	0.5814	7	0.0046	0.0000	0.0006	0.0962	0.7029	0.709	0.0035	0.082	0.2165	0.5893	
		9	-0.0011	0.0001	-0.0001	-0.6218	0.7210	0.0746	0.0032	0.0087														
456	5	7	0.601	0.03	-0.00	0.1157	-0.01	15.14	0.05	15.12	0.2478	0.5684	7	0.0049	0.0001	0.0007	0.1157	0.7128	0.760	0.0037	0.086	0.2100	0.5736	
		9	-0.0007	0.0001	-0.0001	-0.8838	1.0967	0.0796	0.0033	0.0091														
456	6	7	0.601	0.60	-0.00	0.1199	-0.01	15.14	0.05	15.13	0.2421	0.5607	7	0.601	0.0001	0.0006	0.1199	0.6962	0.784	0.0037	0.087	0.2042	0.5628	
		9	-0.0008	0.0001	-0.0001	-0.8810	1.0261	0.0818	0.0033	0.0092														
456	7	7	0.600	1.07	-0.00	0.1353	-0.01	15.14	0.05	15.13	0.2354	0.5555	7	0.600	0.0001	0.0007	0.1353	0.6918	0.801	0.0037	0.089	0.2009	0.5559	
		9	-0.0050	0.0001	-0.0001	-0.9433	1.1041	0.0834	0.0033	0.0092														
456	8	7	0.599	3.17	-0.00	0.1156	-0.01	15.14	0.05	15.12	0.1892	0.5507	7	0.599	0.0001	0.0005	0.1156	0.6332	0.853	0.0032	0.094	0.1614	0.5484	
		9	-0.0009	0.0001	-0.0001	-0.8808	0.7083	0.0888	0.0028	0.0097														
456	9	7	0.600	6.24	-0.00	0.0774	-0.02	15.12	0.05	15.11	0.1486	0.5499	7	0.600	0.0002	0.0004	0.0774	0.7231	0.844	0.0025	0.092	0.1196	0.5474	
		9	-0.0031	0.0000	-0.0000	5.9186	-0.2281	0.0890	0.0021	0.0097														
456	10	7	0.600	9.28	-0.00	0.0772	-0.02	15.12	0.05	15.10	0.1213	0.5547	7	0.600	0.0001	0.0004	0.0772	0.7268	0.895	0.0021	0.10	0.0979	0.5492	
		9	-0.0029	0.0000	-0.0000	-12.0841	3.4488	0.0925	0.0018	0.0101														
456	11	7	0.600	12.44	-0.00	0.0956	-0.02	15.12	0.05	15.10	0.1091	0.5561	7	0.600	0.0001	0.0004	0.0956	0.7424	0.970	0.0020	0.105	0.0886	0.5508	
		9	-0.0031	0.0000	-0.0000	-0.9668	0.9906	0.0970	0.0017	0.0106														
456	12	7	0.599	0.05	-0.00	0.1152	-0.01	15.14	0.05	15.12	0.2473	0.5679	7	0.599	0.0001	0.0006	0.1152	0.7041	0.760	0.0037	0.12	0.2099	0.5713	
		9	-0.0049	0.0001	-0.0001	-0.7438	0.6226	0.0797	0.0033	0.0091														
457	1	7	0.599	-3.01	-0.00	0.0620	-0.02	12.06	0.06	12.07	0.2682	0.6452	7	0.599	0.0000	0.0004	0.0620	0.6968	0.403	0.0021	0.0852	0.2215	0.6406	
		9	-0.0035	0.0000	-0.0002	-0.4227	0.7579	0.0441	0.0019	0.0056														
457	2	7	0.599	-0.97	-0.00	0.1045	-0.01	12.10	0.05	12.10	0.2697	0.6070	7	0.599	0.0000	0.0006	0.1045	0.6857	0.564	0.0030	0.10	0.2202	0.6028	
		9	-0.0009	0.0001	-0.0001	-0.7250	0.7780	0.0599	0.0026	0.0072														

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key															
457	4	7	0.598	0.001	0.02	-0.007	0.1199	-0.001	-0.01	12.11	0.05	12.11	0.0033	0.0029	0.2601	0.0075	0.5914	
		9	-0.0008	0.0001	0.0001	0.0001	-0.8251	0.9275	0.0679	0.0080	0.0029	0.0080	0.0029	0.2166	0.0080	0.5953		
457	4	7	0.600	0.001	0.57	-0.006	0.1264	-0.001	-0.01	12.12	0.06	12.12	0.0034	0.0030	0.2543	0.0079	0.5839	
		9	-0.0007	0.0001	0.0001	0.0001	-0.9498	1.0151	0.0680	0.0083	0.0030	0.0083	0.0030	0.2161	0.0083	0.5876		
457	5	7	0.601	0.001	1.08	-0.007	0.1353	-0.001	-0.02	12.13	0.05	12.12	0.0036	0.0031	0.2531	0.0085	0.5770	
		9	-0.0008	0.0001	0.0001	0.0001	-1.0683	0.7922	0.0740	0.0085	0.0031	0.0085	0.0031	0.2147	0.0085	0.5805		
457	6	7	0.599	0.001	3.16	-0.006	0.1378	-0.001	-0.01	12.13	0.05	12.13	0.0037	0.0033	0.2321	0.0088	0.5512	
		9	-0.0009	0.0001	0.0001	0.0001	-0.9859	0.5976	0.0834	0.0092	0.0033	0.0092	0.0033	0.2013	0.0092	0.5513		
457	7	7	0.601	0.000	6.23	-0.004	0.0727	-0.002	-0.02	12.13	0.06	12.12	0.0029	0.0025	0.1661	0.0096	0.5513	
		9	-0.0001	0.0001	0.0001	0.0000	9.4611	-1.5356	0.0921	0.0100	0.0025	0.0100	0.0025	0.1372	0.0100	0.5468		
457	8	7	0.601	0.000	9.29	-0.004	0.0716	-0.002	-0.02	12.12	0.05	12.11	0.0024	0.0020	0.1446	0.0094	0.5483	
		9	-0.0002	0.0002	0.0002	0.0000	-3.8499	0.1757	0.0894	0.0096	0.0020	0.0096	0.0020	0.1127	0.0096	0.5413		
457	9	7	0.600	0.000	12.46	-0.004	0.0732	-0.001	-0.01	12.12	0.06	12.11	0.0020	0.0017	0.1137	0.0101	0.5569	
		9	-0.0009	0.0001	0.0001	0.0001	-0.7258	0.8319	0.0937	0.0102	0.0017	0.0102	0.0017	0.0913	0.0102	0.5491		
457	10	7	0.599	0.001	0.02	-0.006	0.1144	-0.001	-0.01	12.11	0.05	12.11	0.0033	0.0029	0.2581	0.0080	0.5874	
		9	-0.0008	0.0001	0.0001	0.0001	-0.9454	1.0751	0.0678	0.0080	0.0029	0.0080	0.0029	0.2167	0.0080	0.5929		
458	1	7	0.599	0.000	3.01	-0.004	0.0580	-0.001	-0.02	9.08	0.05	9.01	0.0012	0.0011	0.2680	0.0031	0.6526	
		9	-0.0035	0.0001	0.0001	0.0001	-0.5366	0.7886	0.0242	0.0035	0.0011	0.0035	0.0011	0.2181	0.0035	0.6660		
458	2	7	0.600	0.001	-1.04	-0.006	0.1099	-0.001	-0.01	9.12	0.05	9.05	0.0023	0.0019	0.2726	0.0053	0.6332	
		9	-0.0046	0.0001	0.0001	0.0002	-0.6954	1.3560	0.0446	0.0056	0.0019	0.0056	0.0019	0.2187	0.0056	0.6333		
458	3	7	0.601	0.001	0.03	-0.007	0.1184	-0.001	-0.01	9.14	0.05	9.06	0.0028	0.0023	0.2803	0.0064	0.6089	
		9	-0.0051	0.0001	0.0001	0.0001	-0.8245	0.9554	0.0521	0.0023	0.0023	0.0064	0.0023	0.2266	0.0064	0.6201		

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key															
458	4	7	0.600	0.0001	0.58	-0.0001	0.0006	0.1244	-0.01	9.14	0.0030	0.05	9.06	0.2783	0.6001		
		8	0.0050	0.0001	0.0001	0.0001	0.0006	0.1244	0.6886	0.540	0.0030	0.0030	0.0064	0.2272	0.6103		
		9	-0.0009	0.0001	-0.0001	-0.0001	-0.0001	-0.8257	0.9015	0.0560	0.0025	0.0025	0.0068				
458	5	7	0.600	0.0001	1.04	-0.0001	0.0007	0.1377	-0.01	9.15	0.05	9.07	0.2723	0.5946			
		8	0.0050	0.0001	0.0001	0.0007	0.0007	0.1377	0.6923	0.577	0.0031	0.0031	0.0068	0.2221	0.6029		
		9	-0.0009	0.0001	-0.0001	-0.0001	-0.0001	-0.8785	0.9650	0.0597	0.0026	0.0026	0.0072				
458	6	7	0.601	0.0001	3.15	-0.0001	0.0005	0.1321	-0.01	9.17	0.05	9.10	0.2598	0.5734			
		8	0.0045	0.0001	0.0001	0.0005	0.0005	0.1321	0.6388	0.720	0.0037	0.0037	0.0082	0.2188	0.5772		
		9	-0.0006	0.0001	-0.0001	-0.0001	-0.0001	-1.3541	1.1435	0.0744	0.0032	0.0032	0.0085				
458	7	7	0.600	0.0002	6.22	-0.0002	0.0004	0.0625	-0.02	9.18	0.06	9.10	0.2105	0.5454			
		8	0.0034	0.0002	0.0000	0.0004	0.0004	0.0625	0.6372	0.830	0.0034	0.0034	0.0090	0.1778	0.5421		
		9	-0.0000	0.0002	0.0000	0.0000	-15.4394	-1.1891	0.0869	0.0830	0.0030	0.0030	0.0094				
458	8	7	0.601	0.0001	9.30	-0.0001	0.0004	0.0969	-0.02	9.17	0.05	9.08	0.1553	0.5473			
		8	0.0029	0.0001	0.0000	0.0004	0.0004	0.0969	0.7617	0.855	0.0026	0.0026	0.0093	0.1270	0.5395		
		9	0.0001	0.0001	-0.0001	-0.0000	-0.0000	5.2808	-1.1460	0.0885	0.0022	0.0022	0.0095				
458	9	7	0.600	0.0001	12.42	-0.0001	0.0004	0.0690	-0.02	9.16	0.05	9.08	0.1223	0.5522			
		8	0.0030	0.0001	0.0000	0.0004	0.0004	0.0690	0.7363	0.888	0.0021	0.0021	0.0098	0.0987	0.5451		
		9	-0.0008	0.0001	-0.0001	-0.0001	-0.8317	0.6412	0.7494	0.0913	0.0018	0.0018	0.0099				
458	10	7	0.598	0.0001	0.03	-0.0001	0.0006	0.1235	-0.01	9.13	0.05	9.06	0.2782	0.6051			
		8	0.0049	0.0001	0.0001	0.0006	0.0006	0.1235	0.6982	0.502	0.0027	0.0027	0.0060	0.2251	0.6144		
		9	-0.0009	0.0001	-0.0001	-0.0001	-0.8292	0.7494	0.7494	0.0524	0.0023	0.0023	0.0064				
459	1	7	0.597	-0.0006	-0.0001	0.0002	0.0002	0.3612	3.00	-0.05	0.05	0.02	0.2639	0.3685			
		8	0.0086	-0.0006	-0.0006	0.0010	0.0010	0.3612	0.6122	0.022	0.0001	0.0001	0.0001	0.0048	0.5317		
		9	-0.0009	0.0001	-0.0001	-0.0002	-0.7124	1.0980	0.9072	0.0032	-0.0000	-0.0000	0.0003				
459	2	7	0.599	-0.0003	-0.0001	0.0004	0.0004	0.4755	2.02	-0.05	0.05	0.03	0.3922	0.5434			
		8	0.0040	-0.0003	-0.0003	0.0004	0.0004	0.4755	0.6109	0.022	0.0001	0.0001	0.0001	0.2881	0.3922		
		9	-0.0009	0.0001	-0.0001	-0.0001	-0.7184	0.9072	0.9072	0.0030	-0.0000	-0.0000	0.0003	0.0333	0.5434		
459	3	7	0.599	-0.0001	-0.0001	0.0001	0.0001	-1.2475	-1.03	-0.05	0.05	0.03	0.3342	0.4316			
		8	0.0004	-0.0001	-0.0001	0.0001	0.0001	-1.2475	1.1870	0.022	0.0001	0.0001	0.0001	0.0099	0.4316		
		9	-0.0008	0.0001	-0.0001	-0.0001	-0.8217	1.0736	1.0736	0.0028	-0.0000	-0.0000	0.0003	-0.0099	0.5826		
459	4	7	0.599	-0.0001	-0.0001	0.0003	0.0003	0.0025	-0.53	-0.06	0.05	0.02	0.3551	0.4860			
		8	0.0028	0.0000	0.0000	0.0003	0.0003	0.0025	0.6961	0.022	0.0001	0.0001	0.0001	0.0039	0.4860		
		9	-0.0008	0.0001	-0.0001	-0.0001	-0.7679	1.0006	1.0006	0.0029	-0.0000	-0.0000	0.0003	-0.0039	0.5685		

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key															
459	5	7	0.599	-0.001	-0.006	0.1210	0.6978	-0.02	-0.05	0.05	-0.02	0.0002	0.3550	0.4873				
	6	9	0.0048	0.0001	0.0006	0.1210	0.6978	0.0022	0.0027	0.0001	0.0003	0.0003	0.5462					
			-0.0008	0.0001	-0.0001	-0.8824	1.1257	0.0027	-0.0000	0.0000	0.0000	-0.0066						
459	6	7	0.598	-0.002	-0.010	0.1645	0.6554	0.49	-0.05	0.05	-0.02	0.2923	0.4215					
	6	9	0.0076	0.0001	0.0010	0.1645	0.6554	0.0024	0.0027	0.0001	0.0002	0.0002	0.5312					
			-0.0008	0.0001	-0.0001	-0.8850	0.9389	0.0027	-0.0000	0.0000	0.0000	-0.0130						
459	7	7	0.600	-0.003	-0.013	0.1868	1.02	-0.06	-0.06	0.05	-0.02	0.3321	0.4879					
	6	9	0.0102	0.0001	0.0013	0.1868	1.02	0.0024	0.0027	0.0001	0.0002	0.0003	0.5462					
			-0.0007	0.0001	-0.0001	-0.9521	0.9840	0.0027	-0.0000	0.0000	0.0000	-0.0066						
459	8	7	0.599	-0.006	-0.020	0.2087	2.05	-0.05	-0.05	0.05	-0.02	0.2777	0.4305					
	6	9	0.0153	0.0001	0.0020	0.2087	2.05	0.0027	0.0025	0.0001	0.0002	0.0002	0.5521					
			-0.0006	0.0001	-0.0001	-1.1071	1.1623	0.0025	-0.0000	0.0000	0.0000	-0.0164						
459	9	7	0.599	-0.009	-0.027	0.2226	3.05	-0.06	-0.06	0.05	-0.02	0.3735	0.5281					
	6	9	0.0209	0.0001	0.0027	0.2226	3.05	0.0025	0.0025	0.0001	0.0002	0.0002	0.4814					
			-0.0007	0.0001	-0.0001	-1.0823	0.8796	0.0025	-0.0000	0.0000	0.0000	-0.0479						
460	1	7	0.599	-0.010	-0.032	-0.1190	-0.02	-2.96	-0.0171	0.06	-0.03	0.2973	0.6977					
	6	9	0.0010	0.0002	0.0002	-0.1190	1.3182	-0.0171	0.0016	0.0010	0.0000	0.3510	0.1605					
			-0.0004	0.0002	0.0000	-3.0547	-0.5464	0.0016	-0.0001	0.0001	0.0000	-0.3143	0.7348					
460	2	7	0.597	-0.010	-0.032	-0.0990	-0.02	-1.86	-0.0112	0.06	-0.03	0.3084	0.7348					
	6	9	0.0010	0.0002	0.0002	-0.0990	1.3442	-0.0112	0.0018	0.0006	0.0000	0.3129	0.8060					
			-0.0005	0.0002	0.0000	-2.5992	-0.0117	0.0018	-0.0020	0.0001	0.0000	-0.3129	0.2251					
460	3	7	0.598	-0.015	-0.042	-0.0427	-0.02	-0.95	-0.0064	0.06	-0.03	0.3273	0.8060					
	6	9	0.0015	0.0002	0.0002	-0.0427	0.9652	-0.0064	0.0020	0.0004	0.0010	0.3129	0.9265					
			-0.0006	0.0002	0.0000	-2.1010	-0.1696	0.0020	-0.0001	0.0001	0.0000	-0.3129	0.2050					
460	4	7	0.599	-0.016	-0.043	-0.0196	-0.02	-0.51	-0.0042	0.06	-0.04	0.3841	0.9265					
	6	9	0.0016	0.0002	0.0003	-0.0196	0.9849	-0.0042	0.0020	0.0003	0.0007	0.3128	0.2050					
			-0.0003	0.0002	-0.0000	-3.4914	0.0997	0.0020	-0.0001	0.0001	0.0000	-0.3128	1.1733					
460	5	7	0.599	-0.013	-0.043	-0.0133	-0.02	-0.07	-0.0021	0.05	-0.04	0.4834	1.1733					
	6	9	0.0013	0.0002	0.0003	-0.0133	1.1081	-0.0021	0.0020	0.0002	0.0004	0.3121	0.1740					
			-0.0008	0.0002	-0.0000	-1.3969	0.0356	0.0020	-0.0001	0.0001	0.0000	-0.3121	3.5972					
460	6	7	0.598	-0.014	-0.043	-0.0010	-0.02	0.47	-0.0003	0.05	-0.03	1.3534	3.5972					
	6	9	0.0014	0.0002	0.0003	0.0010	1.1638	-0.0003	0.0020	0.0000	0.0002	0.2565	0.2244					
			-0.0007	0.0002	-0.0000	-1.7397	0.0701	0.0020	-0.0001	0.0001	0.0000	-0.2565						

TABLE I
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key →																							
460	7	0.599	0.0014	0.0007	0.0000	-0.0002	0.0000	89.99	-0.1002	-0.0000	-0.0003	-0.0199	0.0021	0.0001	0.0000	0.0001	1.01	0.0021	-0.0001	0.05	0.0001	-0.0000	0.0001	0.1859	0.2724	0.2116
460	8	0.599	0.0017	0.0008	0.0002	0.0000	89.99	0.0205	0.0003	-0.0003	0.9944	0.0065	0.0021	0.0003	0.0000	0.05	2.09	0.0055	-0.0001	0.05	0.0008	0.0000	0.0000	0.2916	0.6200	0.2240
460	9	0.599	0.0017	0.0007	0.0002	0.0000	90.00	0.0326	0.0003	-0.0000	1.0662	0.0107	0.0023	0.0005	0.05	2.99	0.0023	-0.0001	0.05	0.0013	0.0000	0.0000	0.2636	0.6283	0.2008	
461	1	0.598	0.0001	0.0021	0.0000	0.0000	-89.99	7.4796	-0.0003	-0.0000	1.5948	0.0017	0.0059	0.05	0.05	-0.06	-0.0017	-0.0005	0.05	0.0000	-0.0000	-0.0009	0.1304	0.9606	0.8217	
461	2	0.598	0.0001	0.0020	0.0000	0.0002	-90.00	9.0743	-0.0003	-0.0000	2.5496	0.0018	0.0017	0.04	0.04	-0.05	-0.0017	-0.0003	0.04	0.0000	-0.0004	0.0000	0.1422	0.1103	1.3943	
461	3	0.598	0.0003	0.0022	0.0000	0.0002	-90.00	3.2469	-0.0003	-0.0000	1.0646	0.0018	0.0018	0.05	0.05	-0.05	0.0018	-0.0001	0.05	0.0000	-0.0000	0.0000	0.1422	0.1103	0.1888	
461	4	0.598	0.0002	0.0019	0.0000	0.0002	-90.00	5.0211	-0.0003	-0.0000	2.0632	0.0018	0.0038	0.05	0.05	-0.05	0.0018	0.0000	0.05	0.0000	-0.0003	0.0000	0.1672	0.1737	0.4849	
461	5	0.598	0.0002	0.0018	0.0000	0.0002	-90.00	3.9050	-0.0003	-0.0000	1.2889	0.0018	0.0058	0.05	0.05	-0.05	0.0018	0.0001	0.05	0.0000	-0.0006	0.0000	0.1697	0.1343	0.5843	
461	6	0.598	0.0002	0.0017	0.0000	0.0002	-90.00	4.0383	-0.0003	-0.0001	1.7811	0.0019	0.0061	0.05	0.05	-0.06	0.0019	0.0002	0.05	0.0000	0.0010	0.047	0.1562	0.1617	0.6443	
461	7	0.597	0.0004	0.0018	0.0000	0.0002	-90.00	2.8792	-0.0003	-0.0001	1.2382	0.0020	0.0105	0.05	0.05	-0.06	0.0020	0.0003	0.05	0.0000	0.0013	0.096	0.1641	0.1584	0.6269	
461	8	0.599	0.0005	0.0018	0.0000	0.0002	-90.00	2.2988	-0.0003	-0.0001	1.0432	0.0020	0.0158	0.05	0.05	-0.05	0.0020	0.0006	0.05	0.0000	0.0019	2.01	0.1629	0.1759	0.6053	

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																				
461	9	0.598	0.0006	0.0000	0.0000	0.0000	1.7542	0.7416	-0.003	-0.06	0.05	3.08	0.1723	0.1954	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.2101	0.6206
462	7	1.251	0.0022	0.0000	0.0004	0.0000	-0.2617	0.9128	-0.02	-3.01	0.05	-3.01	0.0789	0.6901	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0814	0.6847
462	8	1.252	0.0029	0.0000	0.0004	0.0000	-0.1569	0.8059	-0.02	-2.98	0.05	-2.98	0.0956	0.7012	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.1235	0.7013
462	9	1.250	0.0030	0.0000	0.0005	0.0000	-0.1223	0.8493	-0.02	-2.96	0.06	-2.97	0.1235	0.7321	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.1697	0.7110
462	10	1.251	0.0030	0.0000	0.0005	0.0000	-0.1223	0.8493	-0.02	-2.95	0.06	-2.97	0.1540	0.7341	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.2277	0.7422
462	11	1.251	0.0036	0.0000	0.0002	0.0000	-0.1310	0.7679	-0.02	-2.95	0.06	-2.96	0.3061	0.9112	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.3441	0.7984
462	12	1.251	0.0033	0.0000	0.0004	0.0000	-0.1366	0.7325	-0.02	-2.92	0.06	-2.94	0.0967	0.6504	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0243	0.6626
462	13	1.251	0.0021	0.0000	0.0003	0.0000	-0.3105	0.7650	-0.02	-2.88	0.06	-2.91	0.0726	0.6660	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0400	0.6654
462	14	1.252	0.0020	0.0000	0.0002	0.0000	-0.2734	0.7750	-0.02	-2.83	0.06	-2.88	0.0670	0.6574	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0407	0.6595
462	15	1.252	0.0014	0.0000	0.0002	0.0000	-0.4149	0.7389	-0.02	-2.80	0.06	-2.85	0.0646	0.6416	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0446	0.6422
462	16	1.252	0.0029	0.0000	0.0002	0.0000	-0.1308	0.8335	-0.02	-2.97	0.06	-2.97	0.1448	0.7487	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.1766	0.7434

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key														
463	1	7	1.251	-3.16	-0.0001	-0.0002	-0.0003	-0.2735	-0.7741	-0.02	-0.316	-0.01	0.06	-1.05	0.0997	0.6999
		6	0.0023	0.0000	0.0000	0.0000	-0.1168	1.2277	-0.0294	-0.0006	-0.0041	-0.0006	-0.0006	-0.0044	0.1078	0.6964
		9	-0.0010	0.0002	-0.0002	-0.0002										
463	2	7	1.251	-1.08	-0.0000	-0.0002	-0.0004	-0.1841	-0.7809	-0.02	-0.98	-0.06	0.06	-1.03	0.1480	0.7186
		6	0.0027	0.0000	0.0000	0.0004	-1.7514	1.7289	-0.0102	-0.0003	-0.0014	-0.0004	-0.0016	-0.0016	0.2065	0.7280
		9	-0.0006	0.0002	-0.0002	-0.0002										
463	3	7	1.251	-0.04	-0.0000	-0.0002	-0.0004	-0.1311	-0.8218	-0.02	-0.96	-0.06	0.06	-1.01	0.3703	0.9344
		6	0.0029	0.0000	0.0000	0.0004	-2.8256	2.1918	-0.0027	-0.0001	-0.0004	-0.0002	-0.0004	-0.0004	0.4675	0.7779
		9	-0.0004	0.0002	-0.0002	-0.0002										
463	4	7	1.252	0.51	-0.0000	-0.0005	-0.1130	-0.8145	-0.095	-0.06	-1.01	0.06	0.06	-0.1969	0.4892	
		6	0.0031	0.0002	0.0000	0.0005	-3.6301	2.5502	0.0014	-0.0001	0.0001	-0.0001	0.0001	0.0001	-0.6283	0.5473
		9	-0.0003	0.0002	-0.0002	-0.0002										
463	5	7	1.251	1.00	-0.0000	-0.0005	-0.1276	-0.7902	-0.094	-0.06	-1.00	0.06	0.06	0.0642	0.6670	
		6	0.0034	0.0000	0.0000	0.0005	-4.5937	3.1575	0.0054	0.0000	0.0000	0.0000	0.0000	0.0007	0.0664	
		9	-0.0002	0.0002	-0.0002	-0.0002								0.0006	0.6460	
463	6	7	1.250	3.13	-0.0000	-0.0004	-0.1333	-0.7573	-0.092	-0.06	-0.98	0.06	0.06	0.0790	0.6559	
		6	0.0032	0.0000	0.0000	0.0004	-3.6757	2.2931	0.0211	0.0002	0.0028	0.0002	0.0030	0.0541	0.6638	
		9	-0.0004	0.0003	-0.0003	-0.0003								0.0541	0.6638	
463	7	7	1.250	6.23	-0.0000	-0.0003	-0.3097	-0.7588	-0.088	-0.06	-0.95	0.06	0.06	0.0546	0.6652	
		6	0.0022	0.0001	0.0000	0.0003	13.2978	-3.4965	0.0507	0.0003	0.0068	0.0003	0.0068	0.0356	0.6679	
		9	-0.0001	0.0003	-0.0003	-0.0003								0.0356	0.6679	
463	8	7	1.251	9.38	-0.0000	-0.0003	-0.3042	-0.7753	-0.084	-0.06	-0.92	0.06	0.06	0.0624	0.6504	
		6	0.0019	0.0001	0.0000	0.0003	-2.6451	1.1783	0.0765	0.0006	0.0101	0.0006	0.0099	0.0385	0.6471	
		9	-0.0006	0.0003	-0.0003	-0.0003								0.0385	0.6471	
463	9	7	1.250	12.61	-0.0000	-0.0002	-0.4149	-0.7389	-0.080	-0.06	-0.89	0.06	0.06	0.0612	0.6327	
		6	0.0014	0.0001	0.0000	0.0002	-1.1527	0.9136	0.0985	0.0012	0.0124	0.0012	0.0124	0.0405	0.6306	
		9	-0.0015	0.0003	-0.0003	-0.0003								0.0405	0.6306	
463	10	7	1.252	-0.03	-0.0000	-0.0004	-0.1406	-0.7818	-0.096	-0.06	-1.01	0.06	0.06	0.0612	1.0770	
		6	0.0030	0.0000	0.0000	0.0004	-2.1490	1.4134	-0.0025	-0.0002	-0.0005	-0.0002	-0.0005	0.5115	0.8606	
		9	-0.0006	0.0002	-0.0002	-0.0002								0.5115	0.8606	
464	1	7	1.251	-3.13	-0.0000	-0.0003	-0.2903	-0.8688	-0.10	-0.06	-0.96	0.05	0.05	0.1169	0.7082	
		6	0.0021	0.0001	0.0000	0.0003	-1.1501	1.0389	-0.0235	-0.0005	-0.0036	-0.0005	-0.0036	0.1270	0.6957	
		9	-0.0012	0.0002	-0.0002	-0.0002								0.1270	0.6957	

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																			
464	2	1.250	-1.06	-0.00	-0.00	-0.02	-0.07	0.06	-0.04	0.2037	0.7520	0.0026	0.0001	0.0004	-0.1908	0.7810	-0.0067	-0.0002	-0.0010	0.2979	0.7494
	6	0.0006	0.0002	-0.0002	0.0004	1.6942	-0.0055	-0.0003	-0.0008												
	9	-0.0006	0.0002	-0.0002	0.0004	-1.6942	-0.0055	-0.0003	-0.0008												
464	3	1.250	-0.03	-0.00	-0.00	-0.02	-0.06	0.05	-0.03	-0.1616	0.4129	0.0031	0.0000	0.0004	-0.1376	0.7881	0.0020	0.0001	0.0001	0.2807	0.7161
	6	0.0004	0.0002	-0.0002	0.0004	2.2034	0.0023	-0.0001	0.0003												
	9	-0.0004	0.0002	-0.0002	0.0004	-2.2034	0.0023	-0.0001	0.0003												
464	4	1.252	0.52	-0.00	-0.00	-0.02	-0.05	0.05	-0.02	0.0672	0.6659	0.0031	0.0000	0.0004	-0.1350	0.7832	0.0054	0.0000	0.0007	0.0386	0.6770
	6	0.0004	0.0002	-0.0002	0.0005	1.9948	0.0054	-0.0000	0.0007												
	9	-0.0004	0.0002	-0.0002	0.0005	-1.9948	0.0054	-0.0000	0.0007												
464	5	1.250	1.03	-0.00	-0.00	-0.01	-0.04	0.05	-0.01	0.0726	0.6511	0.0034	0.0000	0.0004	-0.1405	0.7659	0.0104	0.0001	0.0013	0.0388	0.6988
	6	0.0004	0.0002	-0.0002	0.0005	2.1106	0.0094	0.0000	0.0013												
	9	-0.0004	0.0002	-0.0002	0.0005	-2.1106	0.0094	0.0000	0.0013												
464	6	1.249	3.11	-0.00	-0.00	-0.02	-0.01	0.05	0.00	0.0706	0.6565	0.0036	0.0001	0.0004	-0.1449	0.6791	0.0295	0.0002	0.0036	0.0539	0.6736
	6	0.0036	0.0003	-0.0003	0.0005	3.1799	0.0271	0.0002	0.0036												
	9	-0.0036	0.0003	-0.0003	0.0005	-3.1799	0.0271	0.0002	0.0036												
464	7	1.250	6.26	-0.00	-0.00	-0.02	0.02	0.05	0.03	0.0635	0.6620	0.0024	0.0001	0.0003	-0.2755	0.6778	0.0563	0.0003	0.0076	0.0337	0.6639
	6	0.0024	0.0003	-0.0003	0.0005	6.7438	0.0577	0.0003	0.0076												
	9	-0.0024	0.0003	-0.0003	0.0005	-6.7438	0.0577	0.0003	0.0076												
464	8	1.250	9.41	-0.00	-0.00	-0.02	0.05	0.06	0.05	0.0613	0.6455	0.0021	0.0001	0.0003	-0.2936	0.7120	0.0812	0.0006	0.0108	0.0385	0.6464
	6	0.0021	0.0003	-0.0003	0.0005	9.41	0.0838	0.0006	0.0108												
	9	-0.0021	0.0003	-0.0003	0.0005	-9.41	0.0838	0.0006	0.0108												
464	9	1.250	12.61	-0.00	-0.00	-0.02	0.09	0.06	0.08	0.0600	0.6288	0.0015	0.0001	0.0003	-0.4197	0.6997	0.1020	0.0008	0.0132	0.0400	0.6273
	6	0.0015	0.0003	-0.0003	0.0005	12.61	0.1055	0.0008	0.0132												
	9	-0.0015	0.0003	-0.0003	0.0005	-12.61	0.1055	0.0008	0.0132												
464	10	1.250	-0.02	-0.00	-0.00	-0.02	-0.06	0.05	-0.03	-0.2396	0.5213	0.0030	0.0000	0.0004	-0.1325	0.7749	0.0020	0.0001	0.0002	0.3488	0.5213
	6	0.0030	0.0000	-0.0000	0.0004	0.7749	0.0023	0.0001	0.0002												
	9	-0.0030	0.0000	-0.0000	0.0004	-0.7749	0.0023	0.0001	0.0002												
466	4	1.251	-3.13	-0.00	-0.00	-0.02	0.97	0.05	0.93	0.1016	0.6736	0.0019	0.0001	0.0002	-0.2798	0.9689	-0.0203	0.0004	0.0023	0.1475	0.6674
	6	0.0019	0.0001	-0.0001	0.0003	0.9689	0.0177	0.0005	0.0023												
	9	-0.0019	0.0001	-0.0001	0.0003	-0.9689	0.0177	0.0005	0.0023												
466	5	1.250	-1.06	-0.00	-0.00	-0.02	1.00	0.05	0.95	0.2954	0.7521	0.0026	0.0000	0.0004	-0.1640	0.8721	-0.0012	0.0001	0.0000	0.2613	0.7521
	6	0.0026	0.0000	-0.0000	0.0004	0.8721	0.0002	0.0001	0.0000												
	9	-0.0026	0.0000	-0.0000	0.0004	-0.8721	0.0002	0.0001	0.0000												

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																
466	6	7	1.249	-0.004	-0.000	-0.000	1.002	-0.002	-0.1382	0.8631	0.0061	0.005	0.001	0.000	0.96	0.1617	0.7626		
			0.0030	-0.0000	0.0005	-0.8631	0.0067	0.0001	0.0000	0.0009	0.0220	0.7560							
			0.0001	0.0001	-0.0001	4.8584	-8.0635	0.0067	-0.0001	0.0010									
466	7	6	1.251	0.50	-0.005	-0.005	1.02	-0.02	-0.1235	0.8261	0.0109	0.005	0.000	0.97	0.1324	0.7205			
			0.0032	-0.0000	0.0005	-0.8261	0.0107	0.0001	0.0002	0.0015	0.0505	0.7244							
			0.0001	0.0001	-0.0001	5.5742	-6.7911	0.0107	0.0001	0.0015									
466	8	7	1.250	1.04	-0.005	-0.005	1.03	-0.02	-0.1243	0.7621	0.0150	0.005	0.000	0.97	0.1062	0.6869			
			0.0036	-0.0000	0.0005	-0.7621	0.0150	0.0001	0.0003	0.0022	0.0588	0.6938							
			-0.0000	0.0001	-0.0001	24.2543	0.0150	0.0001	0.0001	0.0020									
466	9	7	1.250	3.15	-0.005	-0.005	1.06	-0.02	-0.1234	0.7075	0.0351	0.005	0.000	0.99	0.0837	0.6828			
			0.0035	-0.0000	0.0005	-0.7075	0.0353	0.0003	0.0005	0.0047	0.0515	0.6866							
			-0.0000	0.0001	-0.0001	27.3971	0.0353	0.0003	0.0003	0.0045									
466	10	7	1.250	6.29	-0.003	-0.003	1.10	-0.02	-0.2995	0.7681	0.0612	0.003	0.000	1.02	0.0727	0.6726			
			0.0023	-0.0001	0.0003	-0.7681	0.0633	0.0004	0.0008	0.0084	0.0369	0.6690							
			0.0005	0.0002	-0.0000	2.0502	0.0633	0.0004	0.0008	0.0084									
466	11	7	1.249	9.39	-0.003	-0.003	1.13	-0.02	-0.2929	0.8557	0.0858	0.003	0.000	1.05	0.0659	0.6497			
			0.0019	-0.0001	0.0003	-0.8557	0.0886	0.0007	0.0011	0.0114	0.0408	0.6458							
			-0.0001	0.0002	-0.0001	4.1517	0.0886	0.0007	0.0011	0.0114									
466	12	7	1.251	12.63	-0.002	-0.002	1.16	-0.02	-0.4037	0.7683	0.1055	0.002	0.000	1.08	0.0634	0.6278			
			0.0015	-0.0001	0.0002	-0.7683	0.1091	0.0008	0.0013	0.0136	0.0405	0.6246							
			-0.0008	0.0002	-0.0002	1.6036	0.1091	0.0008	0.0013	0.0136									
466	13	7	1.251	-0.06	-0.005	-0.005	1.02	-0.02	-0.1247	0.8446	0.0065	0.005	0.000	0.97	0.1328	0.7149			
			0.0029	-0.0000	0.0005	-0.8446	0.0065	0.0001	0.0001	0.0009	0.0044	0.6987							
			-0.0003	0.0002	-0.0001	2.3768	0.0065	0.0001	0.0001	0.0009									
467	1	7	1.252	-3.10	-0.003	-0.003	2.96	-0.02	-0.2735	0.7741	-0.0063	0.003	0.000	3.03	0.1792	0.7082			
			0.0023	-0.0001	0.0003	-0.7741	-0.0060	-0.0003	-0.0011	-0.0008	0.2923	0.6963							
			-0.0007	0.0001	-0.0002	1.4847	-0.0060	-0.0003	-0.0011	-0.0008									
467	2	7	1.249	-1.04	-0.004	-0.004	2.99	-0.02	-0.1609	0.7875	0.0082	0.004	0.000	3.06	0.1353	0.7420			
			0.0028	-0.0000	0.0004	-0.7875	0.0095	0.0002	0.0012	0.0013	0.0414	0.7113							
			-0.0002	0.0001	-0.0002	3.5164	0.0095	0.0002	0.0012	0.0013									
467	3	7	1.251	-0.01	-0.005	-0.005	3.00	-0.02	-0.1169	0.7874	0.0178	0.005	0.000	3.06	0.1045	0.6943			
			0.0031	-0.0000	0.0005	-0.7874	0.0179	0.0003	0.0024	0.0025	0.0609	0.7059							
			-0.0002	0.0001	-0.0002	4.3731	0.0179	0.0003	0.0024	0.0025									

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key												
467	4	7	1.250	0.53	-0.00	-0.005	-0.1308	0.7659	3.02	0.229	0.06	3.07	0.0934	0.6902
		8	0.0034	-0.0000	0.0005	0.0005	-0.1308	0.7659	0.229	0.0004	0.0004	0.0031	0.0591	0.6836
		9	-0.0000	0.0001	-0.0001	-0.0001	-0.1308	0.7659	0.229	0.0002	0.0002	0.0030		
467	5	7	1.251	1.06	-0.00	-0.005	-0.1255	0.7365	3.03	0.283	0.06	3.08	0.0832	0.6866
		8	0.0037	-0.0000	0.0005	0.0005	-0.1255	0.7365	0.283	0.0004	0.0004	0.0038	0.0570	0.6834
		9	-0.0002	0.0002	-0.0001	-0.0001	-0.1255	0.7365	0.283	0.0003	0.0003	0.0037		
467	6	7	1.251	3.16	-0.00	-0.00	-0.1310	0.7534	3.05	0.467	0.06	3.09	0.0750	0.6827
		8	0.0033	-0.0000	0.0005	0.0005	-0.1310	0.7534	0.467	0.0007	0.0007	0.0063	0.0437	0.6825
		9	-0.0001	0.0002	-0.0001	-0.0001	-0.1310	0.7534	0.467	0.0004	0.0004	0.0062		
467	7	7	1.251	6.28	-0.00	-0.00	-0.3192	0.7587	3.09	0.717	0.06	3.13	0.0681	0.6637
		8	0.0022	-0.0001	0.0003	0.0003	-0.3192	0.7587	0.717	0.0009	0.0009	0.0095	0.0370	0.6563
		9	0.0004	0.0002	-0.0001	-0.0001	-0.3192	0.7587	0.717	0.0005	0.0005	0.0098		
467	8	7	1.250	9.45	-0.00	-0.00	-0.3026	0.8650	3.12	0.947	0.06	3.16	0.0622	0.6414
		8	0.0019	-0.0001	0.0003	0.0003	-0.3026	0.8650	0.947	0.0011	0.0011	0.0121	0.0393	0.6375
		9	-0.0003	0.0002	-0.0001	-0.0001	-0.3026	0.8650	0.947	0.0007	0.0007	0.0125		
467	9	7	1.250	12.63	-0.00	-0.00	-0.4191	0.7683	3.14	1.128	0.06	3.18	0.0618	0.6220
		8	0.0015	-0.0001	0.0002	0.0002	-0.4191	0.7683	1.128	0.0013	0.0013	0.0140	0.0405	0.6154
		9	-0.0011	0.0002	-0.0002	-0.0002	-0.4191	0.7683	1.128	0.0009	0.0009	0.0144		
467	10	7	1.249	-0.01	-0.00	-0.00	-2.0089	1.6152	3.01	0.176	0.06	3.07	0.0960	0.6768
		8	0.0031	-0.0000	0.0005	0.0005	-2.0089	1.6152	0.176	0.0003	0.0003	0.0023	0.0223	0.6950
		9	-0.0005	0.0002	-0.0001	-0.0001	-2.0089	1.6152	0.176	0.0002	0.0002	0.0024		
468	1	7	1.250	-3.07	-0.00	-0.00	-0.2446	0.8940	6.01	0.067	0.05	6.00	0.1028	0.7682
		8	0.0022	-0.0001	0.0004	0.0004	-0.2446	0.8940	0.067	0.0001	0.0001	0.0010	0.0286	0.7564
		9	-0.0005	0.0001	-0.0002	-0.0002	-0.2446	0.8940	0.067	0.0000	0.0000	0.0012		
468	2	7	1.251	-1.00	-0.00	-0.00	-2.2488	3.1642	6.04	0.272	0.05	6.03	0.0859	0.6958
		8	0.0028	-0.0000	0.0004	0.0004	-2.2488	3.1642	0.272	0.0004	0.0004	0.0036	0.0563	0.7003
		9	-0.0003	0.0001	-0.0002	-0.0002	-2.2488	3.1642	0.272	0.0003	0.0003	0.0038		
468	3	7	1.251	0.01	-0.00	-0.00	-0.1165	0.7983	6.05	0.363	0.05	6.03	0.0763	0.6907
		8	0.0032	-0.0000	0.0005	0.0005	-0.1165	0.7983	0.363	0.0005	0.0005	0.0050	0.0494	0.6910
		9	-0.0002	0.0001	-0.0002	-0.0002	-0.1165	0.7983	0.363	0.0003	0.0003	0.0050		
468	4	7	1.250	0.55	-0.00	-0.00	-0.1245	0.7399	6.06	0.414	0.05	6.04	0.0746	0.6898
		8	0.0034	-0.0000	0.0005	0.0005	-0.1245	0.7399	0.414	0.0006	0.0006	0.0057	0.0494	0.6923
		9	-0.0001	0.0001	-0.0002	-0.0002	-0.1245	0.7399	0.414	0.0004	0.0004	0.0056		

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																	
468	5	7	1.249	1.008	-0.0001	-0.0002	-0.0001	-0.0005	-0.1322	0.7331	-0.02	6.07	0.06	6.04	0.0724	0.6880	0.0454	0.6840		
		6	0.0036	0.0001	0.0002	0.0001	0.0005	0.0001	-0.4362	0.7331	0.0460	0.0006	0.0063							
		9	-0.0001	0.0002	-0.0001	-0.0001	-0.0001	-0.0001	6.0132	6.0132	0.0455	0.0004	0.0062							
468	6	7	1.250	3.21	-0.0001	-0.0002	-0.0005	-0.1381	0.6820	-0.02	6.10	0.06	6.05	0.0683	0.6705	0.0403	0.6740			
		6	0.0036	0.0002	0.0001	0.0002	0.0005	0.0001	-4.7841	4.0253	0.0636	0.0008	0.0085							
		9	-0.0002	0.0002	-0.0001	-0.0001	-0.0001	-0.0001	0.0635	0.0635	0.0635	0.0005	0.0085							
468	7	7	1.250	6.32	-0.0001	-0.0003	-0.2912	0.7190	-0.02	6.14	0.06	6.09	0.0613	0.6494	0.0384	0.6479				
		6	0.0024	0.0001	0.0002	0.0003	0.0001	0.0001	0.0869	0.0869	0.0869	0.0010	0.0112							
		9	0.0003	0.0002	-0.0001	-0.0001	-0.0001	-0.0001	0.0898	0.0898	0.0898	0.0006	0.0116							
468	8	7	1.250	9.45	-0.0001	-0.0003	-0.3127	0.8385	-0.02	6.16	0.06	6.11	0.0599	0.6295	0.0386	0.6254				
		6	0.0019	0.0001	0.0002	0.0003	0.0001	0.0001	0.1069	0.1069	0.1069	0.0012	0.0134							
		9	-0.0003	0.0002	-0.0001	-0.0001	-0.0001	-0.0001	0.1106	0.1106	0.1106	0.0008	0.0138							
468	9	7	1.252	12.69	-0.0001	-0.0002	-0.4037	0.7683	-0.02	6.19	0.06	6.13	0.0589	0.6146	0.0467	0.6100				
		6	0.0015	0.0001	0.0002	0.0002	0.0002	0.0002	0.1221	0.1221	0.1221	0.0014	0.0150							
		9	-0.0011	0.0002	-0.0001	-0.0001	-0.0001	-0.0001	0.1241	0.1241	0.1241	0.0011	0.0150							
468	10	7	1.251	0.02	-0.0002	-0.0005	-0.1289	0.8157	-0.02	6.06	0.06	6.03	0.0739	0.6857	0.0505	0.6880				
		6	0.0032	0.0001	0.0002	0.0005	0.0001	0.0001	0.2458	0.2458	0.2458	0.0005	0.0049							
		9	-0.0004	0.0002	-0.0001	-0.0001	-0.0001	-0.0001	0.361	0.361	0.361	0.0003	0.0049							
469	3	7	1.250	3.05	-0.0001	-0.0003	-0.2440	0.8879	-0.02	9.08	0.05	9.00	0.0897	0.7159	0.0505	0.7032				
		6	0.0022	0.0001	0.0002	0.0003	0.0001	0.0001	0.0257	0.0257	0.0257	0.0004	0.0034							
		9	-0.0010	0.0002	-0.0001	-0.0001	-0.0001	-0.0001	0.257	0.257	0.257	0.0002	0.0036							
469	4	7	1.249	1.00	-0.0001	-0.0004	-0.1688	0.8701	-0.02	9.11	0.05	9.03	0.0764	0.6866	0.0460	0.6866				
		6	0.0027	0.0000	0.0001	0.0004	0.0001	0.0001	0.0441	0.0441	0.0441	0.0006	0.0061							
		9	-0.0005	0.0001	-0.0001	-0.0001	-0.0001	-0.0001	0.0452	0.0452	0.0452	0.0004	0.0062							
469	5	7	1.249	0.05	-0.0002	-0.0005	-0.1314	0.8213	-0.02	9.12	0.06	9.03	0.0747	0.6900	0.0455	0.6794				
		6	0.0032	0.0000	0.0002	0.0005	0.0001	0.0001	0.0532	0.0532	0.0532	0.0007	0.0073							
		9	-0.0006	0.0002	-0.0001	-0.0001	-0.0001	-0.0001	0.0538	0.0538	0.0538	0.0004	0.0073							
469	6	7	1.250	0.58	-0.0002	-0.0005	-0.1336	0.7523	-0.02	9.13	0.06	9.04	0.0728	0.6847	0.0445	0.6755				
		6	0.0035	0.0000	0.0002	0.0005	0.0000	0.0000	0.0579	0.0579	0.0579	0.0008	0.0079							
		9	-0.0009	0.0002	-0.0001	-0.0001	-0.0001	-0.0001	0.0583	0.0583	0.0583	0.0005	0.0078							
469	7	7	1.250	1.10	-0.0001	-0.0005	-0.1438	0.6978	-0.02	9.14	0.06	9.05	0.0721	0.6791	0.0433	0.6749				
		6	0.0039	0.0001	0.0002	0.0005	0.0000	0.0000	0.0626	0.0626	0.0626	0.0009	0.0085							
		9	-0.0009	0.0002	-0.0001	-0.0001	-0.0001	-0.0001	0.0627	0.0627	0.0627	0.0005	0.0084							

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																							
			1	2	3	4	5	6	7	8	9	10	11	12	13	14	15	16	17	18	19	20	21	22	23	24
469	8	7	1.250	0.0037	-0.00006	3.25	-0.0001	0.00003	-0.0005	0.00000	-2.2534	-0.1426	0.6774	0.5725	9.16	0.0787	0.0792	0.06	0.0010	0.0006	0.0104	0.0103	0.0659	0.0404	0.6612	0.6549
469	9	7	1.249	0.0025	0.00000	6.33	-0.0001	0.00003	-0.0003	0.00000	-2.2347	-0.2749	0.6980	0.5199	9.19	0.0997	0.1033	0.05	0.0012	0.0007	0.0127	0.0130	0.0609	0.0372	0.6415	0.6298
469	10	7	1.249	0.0021	-0.00010	9.49	-0.0001	0.00003	-0.0003	0.00000	-1.5235	-0.3107	0.8212	0.5199	9.22	0.1171	0.1202	0.05	0.0014	0.0010	0.0145	0.0146	0.0614	0.0439	0.6190	0.6085
469	11	7	1.250	0.0016	-0.00015	12.68	-0.0001	0.00003	-0.0002	0.00000	-1.0427	-0.3435	0.8217	0.5856	9.23	0.1287	0.1328	0.06	0.0014	0.0009	0.0155	0.0157	0.0566	0.0370	0.6051	0.5940
469	12	7	1.250	0.0032	-0.00011	0.03	-0.0000	0.00003	-0.0005	0.00000	-1.3525	-0.1289	0.8157	0.4718	9.12	0.0530	0.0531	0.06	0.0007	0.0004	0.0072	0.0071	0.0692	0.0451	0.6823	0.6764
470	1	7	1.250	0.0023	-0.00014	3.03	-0.0001	0.00002	-0.0003	0.00000	-0.9146	-0.2644	0.8036	0.6510	12.04	0.0418	0.0435	0.06	0.0005	0.0003	0.0060	0.0060	0.0680	0.0428	0.6972	0.6921
470	2	7	1.250	0.0028	-0.00010	-0.97	-0.0000	0.00002	-0.0001	0.00000	-1.2521	-0.1613	0.7752	0.7375	12.07	0.0607	0.0622	0.05	0.0008	0.0005	0.0082	0.0084	0.0680	0.0424	0.6820	0.6761
470	3	7	1.249	0.0032	-0.00009	0.04	-0.0000	0.00002	-0.0005	0.00000	-1.5504	-0.1478	0.7897	0.7658	12.08	0.0691	0.0704	0.06	0.0009	0.0005	0.0093	0.0093	0.0685	0.0409	0.6732	0.6667
470	4	7	1.250	0.0035	-0.00011	0.58	-0.0001	0.00003	-0.0005	0.00000	-1.3627	-0.1511	0.7093	0.5497	12.09	0.0734	0.0743	0.06	0.0009	0.0006	0.0098	0.0098	0.0679	0.0412	0.6675	0.6626
470	5	7	1.251	0.0038	-0.00011	1.12	-0.0001	0.00003	-0.0005	0.00000	-1.3983	-0.1371	0.7146	0.5212	12.10	0.0778	0.0785	0.06	0.0010	0.0006	0.0103	0.0103	0.0670	0.0400	0.6639	0.6567
470	6	7	1.250	0.0036	-0.00008	3.22	-0.0000	0.00003	-0.0005	0.00000	-2.0607	-0.1354	0.6941	0.6174	12.12	0.0928	0.0940	0.06	0.0011	0.0007	0.0120	0.0120	0.0602	0.0373	0.6473	0.6430

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																		
470	7	1.250	6.55	-0.00	-0.02	12.14	0.06	12.13	0.0588	0.6273	0.0024	-0.0001	0.0003	0.2655	-0.7681	0.1113	0.0013	0.0139	0.0579	0.6168
	8	0.0024	-0.0001	0.0003	0.0000	-11.7310	-0.2646	0.1151	0.0008	0.0142	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
	9	-0.0001	0.0003	0.0000	0.0000	-0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
470	7	1.248	9.49	-0.00	-0.02	12.16	0.06	12.14	0.0572	0.6108	0.0023	-0.0001	0.0003	0.2749	-0.6998	0.1252	0.0014	0.0152	0.0424	0.5946
	8	0.0023	-0.0001	0.0003	0.0000	-1.9447	0.4418	0.1269	0.0010	0.0150	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
	9	-0.0008	0.0003	0.0000	0.0000	-0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
470	7	1.250	12.69	-0.00	-0.02	12.17	0.06	12.15	0.0559	0.5976	0.0017	-0.0001	0.0003	0.3770	-0.7643	0.1345	0.0015	0.0160	0.0370	0.5876
	8	0.0017	-0.0001	0.0003	0.0000	-1.1188	0.7315	0.1391	0.0010	0.0163	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
	9	-0.0015	0.0003	0.0000	0.0000	-0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
470	7	1.249	0.03	-0.00	-0.02	12.08	0.06	12.08	0.0665	0.6700	0.0032	-0.0000	0.0003	0.1508	-0.7636	0.0689	0.0009	0.0092	0.0398	0.6629
	8	0.0032	-0.0000	0.0003	0.0000	-1.4194	0.4363	0.0700	0.0005	0.0092	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
	9	-0.0012	0.0003	0.0000	0.0000	-0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
471	7	1.250	3.01	-0.00	-0.02	15.08	0.06	15.06	0.0625	0.6807	0.0024	-0.0001	0.0003	0.2825	-0.7338	0.0592	0.0007	0.0080	0.0389	0.6791
	8	0.0024	-0.0001	0.0003	0.0000	-1.0082	0.7370	0.0611	0.0004	0.0083	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
	9	-0.0013	0.0002	0.0000	0.0000	-0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
471	7	1.248	0.94	-0.00	-0.02	15.11	0.06	15.08	0.0646	0.6654	0.0028	-0.0001	0.0003	0.1764	-0.7461	0.0769	0.0009	0.0102	0.0397	0.6613
	8	0.0028	-0.0001	0.0003	0.0000	-1.1956	0.6966	0.0781	0.0006	0.0103	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
	9	-0.0011	0.0002	0.0000	0.0000	-0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
471	7	1.251	0.08	-0.00	-0.02	15.12	0.06	15.08	0.0653	0.6564	0.0032	-0.0000	0.0003	0.1245	-0.7937	0.0845	0.0011	0.0111	0.0391	0.6546
	8	0.0032	-0.0000	0.0003	0.0000	-1.2469	0.5611	0.0855	0.0006	0.0112	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
	9	-0.0013	0.0003	0.0000	0.0000	-0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
471	7	1.248	0.59	-0.00	-0.02	15.13	0.06	15.09	0.0646	0.6522	0.0033	-0.0000	0.0003	0.1273	-0.7890	0.0883	0.0011	0.0115	0.0378	0.6487
	8	0.0033	-0.0000	0.0003	0.0000	-1.3673	0.5432	0.0895	0.0006	0.0116	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
	9	-0.0011	0.0003	0.0000	0.0000	-0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
471	7	1.251	1.13	-0.00	-0.02	15.13	0.06	15.09	0.0628	0.6474	0.0037	-0.0001	0.0003	0.1398	-0.6984	0.0922	0.0011	0.0119	0.0379	0.6456
	8	0.0037	-0.0001	0.0003	0.0000	-1.2651	0.5604	0.0932	0.0007	0.0120	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
	9	-0.0012	0.0003	0.0000	0.0000	-0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
471	7	1.250	3.24	-0.00	-0.02	15.15	0.06	15.11	0.0590	0.6323	0.0035	-0.0000	0.0003	0.1382	-0.6765	0.1057	0.0012	0.0133	0.0374	0.6281
	8	0.0035	-0.0000	0.0003	0.0000	-1.6328	0.5500	0.1069	0.0008	0.0134	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
	9	-0.0010	0.0003	0.0000	0.0000	-0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
471	7	1.250	6.38	-0.00	-0.02	15.17	0.06	15.13	0.0617	0.6139	0.0023	-0.0001	0.0003	0.2569	-0.7593	0.1205	0.0014	0.0148	0.0481	0.6052
	8	0.0023	-0.0001	0.0003	0.0000	-0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
	9	0.0000	0.0003	0.0000	0.0000	25.2150	-0.3049	0.1230	0.0011	0.0148	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key											
471	8	7	1.250	9.48	-0.003	-0.00	-0.02	15.18	0.06	15.13	0.0567	0.5988		
		8	0.0024	-0.0001	0.0000	0.0003	0.6769	0.1307	0.0014	0.0156	0.0374	0.5931		
		9	-0.0012	0.0003	-0.0000	-0.0003	0.3737	0.1349	0.0010	0.0160				
471	9	7	1.251	12.68	-0.002	-0.00	-0.02	15.18	0.06	15.14	0.0507	0.5903		
		8	0.0017	-0.0001	0.0002	0.0002	0.6886	0.1388	0.0014	0.0163	0.0357	0.5814		
		9	-0.0020	0.0003	-0.0002	-0.0002	0.7001	0.1411	0.0010	0.0164				
471	10	7	1.250	0.08	-0.00	-0.00	-0.02	15.11	0.05	15.08	0.0632	0.6527		
		8	0.0033	-0.0000	0.0005	0.0005	0.7742	0.0844	0.0010	0.0110	0.0387	0.6521		
		9	-0.0014	0.0003	-0.0001	-0.0001	0.4802	0.0854	0.0006	0.0111				
472	1	7	1.046	-2.99	-0.00	-0.00	-0.02	15.09	0.05	15.06	0.0761	0.6760		
		8	0.0032	-0.0001	0.0004	0.0004	0.7073	0.0678	0.0010	0.0091	0.0480	0.6685		
		9	-0.0011	0.0002	-0.0001	-0.0001	0.7597	0.0707	0.0006	0.0094				
472	2	7	1.048	-0.94	-0.00	-0.00	-0.02	15.12	0.06	15.08	0.0756	0.6527		
		8	0.0043	-0.0001	0.0005	0.0005	0.6567	0.0865	0.0013	0.0113	0.0498	0.6525		
		9	-0.0007	0.0002	-0.0001	-0.0001	0.7519	0.0890	0.0008	0.0116				
472	3	7	1.046	0.08	-0.00	-0.00	-0.02	15.13	0.06	15.09	0.0769	0.6418		
		8	0.0047	-0.0000	0.0006	0.0006	0.7023	0.0947	0.0014	0.0121	0.0487	0.6431		
		9	-0.0006	0.0002	-0.0000	-0.0000	0.7671	0.0973	0.0009	0.0125				
472	4	7	1.047	0.59	-0.00	-0.00	-0.02	15.13	0.06	15.09	0.0817	0.6352		
		8	0.0051	-0.0000	0.0006	0.0006	0.6577	0.0977	0.0015	0.0128	0.0506	0.6383		
		9	-0.0006	0.0002	-0.0000	-0.0000	0.4492	0.1007	0.0010	0.0128				
472	8	7	1.048	1.09	-0.00	-0.00	-0.02	15.14	0.05	15.11	0.0919	0.6346		
		8	0.0053	-0.0000	0.0007	0.0007	0.6806	0.0998	0.0018	0.0126	0.0586	0.6364		
		9	0.0001	0.0001	-0.0000	-0.0000	-2.6385	0.1029	0.0012	0.0131				
472	9	7	1.048	3.20	-0.00	-0.00	-0.02	15.18	0.05	15.12	0.0962	0.6151		
		8	0.0046	-0.0000	0.0006	0.0006	0.6886	0.1097	0.0021	0.0135	0.0754	0.6118		
		9	0.0002	0.0001	-0.0000	-0.0000	-1.4851	0.1121	0.0016	0.0137				
472	10	7	1.049	6.31	-0.00	-0.00	-0.02	15.17	0.05	15.14	0.0843	0.6012		
		8	0.0031	-0.0000	0.0004	0.0004	0.6941	0.1235	0.0020	0.0148	0.0622	0.5942		
		9	0.0005	0.0002	0.0000	0.0000	0.0291	0.1277	0.0015	0.0151				
472	11	7	1.047	9.42	-0.00	-0.00	-0.02	15.17	0.05	15.14	0.0704	0.5926		
		8	0.0030	-0.0000	0.0004	0.0004	0.7227	0.1346	0.0018	0.0159	0.0476	0.5863		
		9	-0.0001	0.0001	-0.0000	-0.0000	2.4689	0.1386	0.0013	0.0162				

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key													
472	12	7	1.046	12.61	-0.004	-0.00	-0.02	15.18	0.06	15.14	0.0585	0.5893				
		8	0.0029	-0.0000	0.0004	0.0004	0.7214	0.1417	0.0016	0.0167	0.0380	0.5805				
		9	-0.0009	0.0001	-0.0001	-0.0001	1.0027	0.1451	0.0011	0.0168						
472	13	7	1.049	0.06	-0.00	-0.02	15.12	0.05	15.10	0.0813	0.6481					
		8	0.0051	-0.0000	0.0006	0.6567	0.0935	0.0015	0.0121	0.0484	0.6461					
		9	-0.0001	0.0001	-0.0000	1.9441	0.0971	0.0009	0.0125							
473	1	7	1.047	-3.02	-0.00	-0.02	12.05	0.05	12.05	0.0928	0.6967					
		8	0.0031	-0.0000	0.0005	0.8166	0.0474	0.0008	0.0066	0.0569	0.6918					
		9	-0.0005	0.0001	-0.0001	1.7406	0.0506	0.0005	0.0070							
473	2	7	1.048	-0.97	-0.00	-0.01	12.07	0.05	12.07	0.0852	0.6829					
		8	0.0043	-0.0000	0.0006	0.7037	0.0676	0.0011	0.0092	0.0533	0.6742					
		9	-0.0002	0.0001	-0.0001	2.4710	0.0708	0.0007	0.0095							
473	3	7	1.047	0.05	-0.00	-0.02	12.08	0.05	12.08	0.0840	0.6690					
		8	0.0049	-0.0000	0.0006	0.6938	0.0775	0.0013	0.0103	0.0535	0.6627					
		9	-0.0002	0.0002	-0.0000	1.2028	0.0796	0.0008	0.0105							
473	4	7	1.050	0.60	-0.00	-0.02	12.10	0.05	12.08	0.0838	0.6617					
		8	0.0051	-0.0000	0.0006	0.6697	0.0822	0.0013	0.0108	0.0520	0.6584					
		9	-0.0001	0.0001	-0.0000	2.2441	0.0842	0.0008	0.0110							
473	5	7	1.047	1.10	-0.00	-0.02	12.10	0.04	12.08	0.0851	0.6527					
		8	0.0051	-0.0000	0.0006	0.6781	0.0862	0.0014	0.0113	0.0525	0.6527					
		9	-0.0000	0.0001	-0.0000	3.9064	0.0884	0.0009	0.0115							
473	6	7	1.047	3.21	-0.00	-0.02	12.12	0.04	12.10	0.0915	0.6331					
		8	0.0043	-0.0000	0.0006	0.7103	0.1004	0.0018	0.0127	0.0634	0.6286					
		9	0.0000	0.0002	-0.0000	-9.0786	0.1028	0.0013	0.0129							
473	7	7	1.052	6.31	-0.00	-0.02	12.14	0.05	12.12	0.0877	0.5986					
		8	0.0032	-0.0000	0.0004	0.6948	0.1136	0.0019	0.0138	0.0681	0.5986					
		9	0.0004	0.0002	0.0000	0.2041	0.1175	0.0016	0.0140							
473	8	7	1.049	9.41	-0.00	-0.02	12.16	0.05	12.13	0.0808	0.5955					
		8	0.0029	-0.0000	0.0004	0.7135	0.1275	0.0020	0.0151	0.0576	0.5861					
		9	-0.0001	0.0002	-0.0000	3.2059	0.1316	0.0015	0.0154							
473	9	7	1.045	12.59	-0.00	-0.02	12.16	0.05	12.13	0.0636	0.5916					
		8	0.0030	-0.0000	0.0004	0.7100	0.1401	0.0017	0.0165	0.0421	0.5816					
		9	-0.0007	0.0002	-0.0001	0.8684	0.1440	0.0012	0.0167							

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																				
473	10	7	1.048	0.06	-0.00	-0.0522	-0.02	12.04	0.06	12.08	0.0830	0.6669	0.0049	-0.0000	0.0006	-0.0522	-0.02	0.6674	0.012	0.0103	0.0830	0.6669	
		8	-0.0004	0.0002	-0.0000	-2.8494	0.5591	0.0775	0.0008	0.0104	0.0531	0.651											
474	1	7	1.051	-3.03	-0.00	-0.1495	-0.02	9.08	0.05	9.01	0.1089	0.6980	0.0033	-0.0001	0.0005	-0.1495	-0.02	0.751	0.006	0.0038	0.1089	0.6980	
		8	-0.0005	0.0001	-0.0001	-1.6288	1.7863	0.0276	0.0004	0.0041	0.0669	0.6912											
474	2	7	1.050	-1.01	-0.00	-0.0902	-0.02	9.11	0.05	9.03	0.0952	0.6879	0.0042	-0.0000	0.0006	-0.0902	-0.02	0.731	0.009	0.0067	0.0952	0.6879	
		8	-0.0005	0.0002	-0.0001	-1.9594	1.1843	0.0494	0.0006	0.0069	0.0607	0.6852											
474	3	7	1.049	0.03	-0.00	-0.0513	-0.02	9.13	0.06	9.04	0.0904	0.6809	0.0049	-0.0000	0.0006	-0.0513	-0.02	0.6916	0.010	0.0081	0.0904	0.6809	
		8	-0.0004	0.0002	-0.0000	-2.5756	0.7424	0.0597	0.0006	0.0082	0.0559	0.6781											
474	4	7	1.048	0.55	-0.00	-0.0485	-0.02	9.13	0.05	9.04	0.0885	0.6768	0.0051	-0.0000	0.0006	-0.0485	-0.02	0.6746	0.011	0.0088	0.0885	0.6768	
		8	-0.0002	0.0002	-0.0000	-4.0392	1.5387	0.0650	0.0007	0.0088	0.0555	0.6731											
474	5	7	1.049	1.06	-0.00	-0.0386	-0.02	9.14	0.05	9.05	0.0885	0.6723	0.0052	-0.0000	0.0007	-0.0386	-0.02	0.6839	0.012	0.0094	0.0885	0.6723	
		8	-0.0003	0.0002	-0.0000	-3.2352	0.9132	0.0701	0.0008	0.0094	0.0569	0.6719											
474	6	7	1.050	3.19	-0.00	-0.0453	-0.02	9.16	0.05	9.06	0.0868	0.6520	0.0045	-0.0000	0.0006	-0.0453	-0.02	0.6843	0.015	0.0114	0.0868	0.6520	
		8	-0.0000	0.0002	-0.0000	-18.0920	5.4051	0.0875	0.0009	0.0115	0.0548	0.6490											
474	7	7	1.047	6.31	-0.00	-0.1432	-0.02	9.19	0.05	9.10	0.0974	0.6217	0.0030	-0.0000	0.0004	-0.1432	-0.02	0.7223	0.020	0.0130	0.0974	0.6217	
		8	-0.0006	0.0002	-0.0000	2.1144	-0.0634	0.1052	0.0016	0.0131	0.0760	0.6099											
474	8	7	1.049	9.42	-0.00	-0.1275	-0.02	9.21	0.05	9.10	0.0866	0.6011	0.0031	-0.0000	0.0004	-0.1275	-0.02	0.7321	0.020	0.0143	0.0866	0.6011	
		8	-0.0001	0.0002	-0.0000	-7.0355	1.2642	0.1196	0.0015	0.0146	0.0644	0.5945											
474	9	7	1.049	12.59	-0.00	-0.1505	-0.02	9.22	0.05	9.12	0.0724	0.5925	0.0029	-0.0000	0.0004	-0.1505	-0.02	0.6949	0.019	0.0158	0.0724	0.5925	
		8	-0.0003	0.0002	-0.0001	-1.2062	1.1110	0.1355	0.0014	0.0161	0.0514	0.5862											
474	10	7	1.052	0.05	-0.00	-0.0483	-0.02	9.12	0.06	9.04	0.0878	0.6772	0.0049	-0.0000	0.0006	-0.0483	-0.02	0.6856	0.010	0.0080	0.0878	0.6772	
		8	-0.0004	0.0002	-0.0000	-2.7856	0.9879	0.0597	0.0006	0.0082	0.0562	0.6773											

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key													
475	1	7	1.048	-0.0001	-0.0002	-0.0005	-0.1472	-0.7281	0.0106	6.01	0.0002	0.05	6.00	0.1174	0.6934
		8	0.0036	0.0002	0.0002	0.0002	-1.6193	1.5981	0.0106	0.0088	0.0000	0.0012	0.0014	0.0442	0.6936
		9	-0.0006	0.0002	-0.0002	-0.0002	-0.0002	1.5981	0.0106	0.0088	0.0000	0.0012	0.0014	0.0442	0.6936
475	2	7	1.050	-1.0000	-0.0002	-0.0006	-0.0848	-0.7509	0.0300	6.04	0.0006	0.06	6.02	0.1205	0.6833
		8	0.0043	0.0002	0.0002	0.0002	-3.9398	2.9460	0.0300	0.287	0.0004	0.0039	0.0041	0.0770	0.6852
		9	-0.0002	0.0002	-0.0002	-0.0002	-0.0002	2.9460	0.0300	0.287	0.0004	0.0039	0.0041	0.0770	0.6852
475	3	7	1.049	0.0100	-0.0002	-0.0006	-0.0514	-0.6933	0.0400	6.05	0.0008	0.05	6.03	0.1127	0.6808
		8	0.0049	0.0000	0.0002	0.0002	-10.5096	7.0153	0.0400	0.396	0.0005	0.0054	0.0054	0.0747	0.6816
		9	-0.0000	0.0002	-0.0002	-0.0002	10.5096	7.0153	0.0400	0.396	0.0005	0.0054	0.0054	0.0747	0.6816
475	5	7	1.049	1.0600	-0.0002	-0.0007	-0.0424	-0.6668	0.0508	6.07	0.0010	0.05	6.05	0.1013	0.6817
		8	0.0053	0.0002	0.0002	0.0002	-2.9039	1.2643	0.0508	0.508	0.0006	0.0069	0.0068	0.0662	0.6781
		9	-0.0003	0.0002	-0.0002	-0.0002	2.9039	1.2643	0.0508	0.508	0.0006	0.0069	0.0068	0.0662	0.6781
475	6	7	1.053	0.0000	-0.0002	-0.0006	-0.0457	-0.6770	0.0709	6.10	0.0012	0.06	6.07	0.0912	0.6655
		8	0.0044	0.0002	0.0002	0.0002	-5.2151	1.5507	0.0709	0.713	0.0008	0.0095	0.0095	0.0586	0.6662
		9	-0.0002	0.0002	-0.0002	-0.0002	5.2151	1.5507	0.0709	0.713	0.0008	0.0095	0.0095	0.0586	0.6662
475	7	7	1.049	0.0000	-0.0002	-0.0004	-0.1374	-0.7025	0.0989	6.12	0.0016	0.06	6.09	0.0879	0.6392
		8	0.0032	0.0002	0.0002	0.0002	-1.4844	0.2209	0.0989	0.945	0.0011	0.0125	0.0125	0.0591	0.6340
		9	-0.0008	0.0002	-0.0002	-0.0002	1.4844	0.2209	0.0989	0.945	0.0011	0.0125	0.0125	0.0591	0.6340
475	8	7	1.049	0.0000	-0.0002	-0.0004	-0.1497	-0.7218	0.1112	6.15	0.0020	0.06	6.12	0.0916	0.6102
		8	0.0029	0.0002	0.0002	0.0002	-3.4802	0.7639	0.1112	1.144	0.0015	0.0138	0.0138	0.0694	0.6034
		9	-0.0003	0.0002	-0.0002	-0.0002	3.4802	0.7639	0.1112	1.144	0.0015	0.0138	0.0138	0.0694	0.6034
475	9	7	1.047	12.5900	-0.0002	-0.0004	-0.1507	-0.6632	0.1260	6.16	0.0020	0.05	6.13	0.0823	0.5957
		8	0.0031	0.0002	0.0002	0.0002	-1.1916	0.9972	0.1300	1.300	0.0015	0.0153	0.0153	0.0601	0.5887
		9	-0.0008	0.0002	-0.0002	-0.0002	1.1916	0.9972	0.1300	1.300	0.0015	0.0153	0.0153	0.0601	0.5887
475	10	7	1.048	0.0000	-0.0002	-0.0006	-0.0654	-0.6638	0.0397	6.05	0.0008	0.06	6.04	0.1108	0.6783
		8	0.0049	0.0002	0.0002	0.0002	-2.6717	1.1312	0.0397	0.400	0.0005	0.0053	0.0053	0.0736	0.6773
		9	-0.0004	0.0002	-0.0002	-0.0002	2.6717	1.1312	0.0397	0.400	0.0005	0.0053	0.0053	0.0736	0.6773
476	1	7	1.050	-0.0001	-0.0002	-0.0005	-0.1553	-0.7427	0.0070	2.95	0.0004	0.05	3.03	0.2925	0.6948
		8	0.0035	0.0002	0.0002	0.0002	-2.0621	1.7051	-0.0048	0.070	-0.0003	-0.0006	-0.0006	0.3942	0.6608
		9	-0.0005	0.0002	-0.0002	-0.0002	2.0621	1.7051	-0.0048	0.070	-0.0003	-0.0006	-0.0006	0.3942	0.6608
476	2	7	1.048	-1.0600	-0.0002	-0.0006	-0.1025	-0.7225	0.0093	2.99	0.0002	0.06	3.05	0.1592	0.6789
		8	0.0044	0.0002	0.0002	0.0002	-4.5011	2.6077	0.0093	0.110	0.0001	0.0012	0.0012	0.0662	0.6758
		9	-0.0002	0.0002	-0.0002	-0.0002	4.5011	2.6077	0.0093	0.110	0.0001	0.0012	0.0012	0.0662	0.6758

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key												
476	3	7	1.048	-0.0002	-0.0000	-0.0006	-0.0001	-0.0615	-0.001	3.00	0.0005	3.07	0.1458	0.6764
		8	0.0048	0.0000	0.0006	0.0006	0.0001	-0.0615	0.6902	0.0190	0.0003	0.0025	0.0784	0.6727
		9	-0.0002	0.0002	-0.0001	-0.0001	-0.0001	-5.1425	3.3233	0.0196	0.0003	0.0026		
476	4	7	1.051	0.51	0.00	0.00	0.00	-0.0618	-0.02	3.01	0.05	3.07	0.1362	0.6708
		8	0.0048	-0.0000	0.0006	0.0006	0.0001	-0.0618	0.6820	0.0247	0.0005	0.0033	0.0795	0.6675
		9	-0.0002	0.0002	-0.0001	-0.0001	-0.0001	-5.3256	3.3146	0.0248	0.0003	0.0033		
476	5	7	1.047	1.04	0.00	0.00	0.00	-0.0522	-0.02	3.01	0.05	3.08	0.1297	0.6688
		8	0.0051	-0.0002	0.0006	0.0006	0.0001	-0.0522	0.6574	0.0302	0.0007	0.0040	0.0833	0.6717
		9	-0.0002	0.0002	-0.0001	-0.0001	-0.0001	-4.7747	2.5401	0.0296	0.0004	0.0039		
476	6	7	1.047	3.15	0.00	0.00	0.00	-0.0457	-0.02	3.04	0.06	3.10	0.1047	0.6738
		8	0.0044	-0.0000	0.0006	0.0006	0.0001	-0.0457	0.6770	0.0519	0.0010	0.0070	0.0728	0.6745
		9	-0.0000	0.0002	-0.0000	-0.0000	-0.0000	-17.1640	7.6503	0.0514	0.0007	0.0069		
476	7	7	1.049	6.26	0.00	0.00	0.00	-0.1532	-0.02	3.08	0.06	3.12	0.0904	0.6527
		8	0.0032	-0.0000	0.0004	0.0004	0.0000	-0.1532	0.6729	0.0790	0.0014	0.0103	0.0599	0.6522
		9	0.0006	0.0002	0.0000	0.0000	0.0000	2.0560	0.4789	0.0829	0.0009	0.0108		
476	8	7	1.046	9.39	0.00	0.00	0.00	-0.1563	-0.02	3.11	0.06	3.14	0.0880	0.6279
		8	0.0031	-0.0000	0.0004	0.0004	0.0000	-0.1563	0.6878	0.1028	0.0018	0.0129	0.0650	0.6185
		9	0.0001	0.0002	-0.0000	-0.0000	-0.0000	10.0426	-3.0974	0.1060	0.0013	0.0131		
476	9	7	1.050	12.56	0.00	0.00	0.00	-0.1918	-0.02	3.13	0.05	3.16	0.0812	0.6058
		8	0.0030	-0.0001	0.0004	0.0004	0.0001	-0.1918	0.6674	0.1191	0.0019	0.0144	0.0633	0.5965
		9	-0.0011	0.0002	-0.0001	-0.0001	-0.0001	-0.9454	0.7928	0.1211	0.0015	0.0144		
476	10	7	1.047	-0.02	0.00	0.00	0.00	-0.0548	-0.02	3.00	0.05	3.06	0.1410	0.6652
		8	0.0047	-0.0000	0.0006	0.0006	0.0001	-0.0548	0.7031	0.0191	0.0005	0.0025	0.0794	0.6647
		9	-0.0005	0.0002	-0.0001	-0.0001	-0.0001	-2.5035	1.0348	0.0197	0.0003	0.0026		
477	1	7	1.049	-3.09	0.00	0.00	0.00	-0.1531	-0.02	0.97	0.05	0.92	0.1700	0.6681
		8	0.0034	-0.0001	0.0005	0.0005	0.0001	-0.1531	0.7352	-0.0199	0.0006	0.0026	0.1978	0.6801
		9	-0.0008	0.0002	-0.0001	-0.0001	-0.0001	-1.5327	1.1678	-0.0172	-0.0006	-0.0023		
477	2	7	1.048	-1.03	0.00	0.00	0.00	-0.1077	-0.02	1.00	0.06	0.95	-1.4847	1.2413
		8	0.0044	-0.0000	0.0006	0.0006	0.0001	-0.1077	0.6860	-0.0003	0.0000	0.0000	-5.5854	-0.5323
		9	-0.0003	0.0002	-0.0001	-0.0001	-0.0001	-3.4116	2.4631	0.0001	-0.0001	-0.0000		
478	3	7	1.045	-3.09	0.00	0.00	0.00	-0.2020	-0.02	0.97	0.05	0.92	0.1560	0.6452
		8	0.0032	-0.0001	0.0004	0.0004	0.0001	-0.2020	0.7328	-0.0203	0.0006	0.0026	0.2002	0.6756
		9	-0.0006	0.0001	-0.0001	-0.0001	-0.0001	-1.3417	1.4834	-0.0169	-0.0006	-0.0022		

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

← See Key →

Run Pt.	Cd																						
478	4	1.0047	-1.0044	-0.0001	-0.0006	-0.1215	0.7275	-0.091	1.000	-0.0010	0.066	-0.0000	0.95	2.9317	0.3630								
	8	0.0044	-0.0001	-0.0001	0.0006	-0.1215	0.7275	-0.091	1.000	-0.0010	0.066	-0.0000	0.95	2.9317	0.3630								
	9	0.0001	-0.0001	-0.0001	-0.0001	5.0644	-4.0815	-0.0010	-0.0010	-0.0001	-0.0001	-0.0000	0.0000	-0.8907	0.3313								
478	5	1.0045	-0.0001	-0.0001	0.0007	-0.0702	0.7802	-0.002	1.023	0.0079	0.055	0.0009	0.96	0.2019	0.6007								
	8	0.0045	-0.0001	-0.0001	0.0007	-0.0702	0.7802	-0.002	1.023	0.0079	0.055	0.0009	0.96	0.2019	0.6007								
	9	0.0001	-0.0001	-0.0001	-0.0001	8.0237	-5.1323	0.0079	0.0079	0.0000	0.0000	0.0010	0.0010	0.0479	0.6747								
478	6	1.0048	-0.0001	-0.0001	0.0007	-0.0603	0.7331	-0.011	1.023	0.0118	0.066	0.0015	0.96	0.1853	0.6067								
	8	0.0049	-0.0001	-0.0001	0.0007	-0.0603	0.7331	-0.011	1.023	0.0118	0.066	0.0015	0.96	0.1853	0.6067								
	9	0.0000	-0.0001	-0.0001	-0.0001	18.4558	-11.3377	0.0118	0.0118	0.0004	0.0004	0.0015	0.0015	0.0742	0.6481								
478	7	1.0046	-0.0001	-0.0001	0.0007	-0.0441	0.7105	-0.010	1.003	0.0171	0.066	0.0022	0.97	0.1635	0.6692								
	8	0.0049	-0.0001	-0.0001	0.0007	-0.0441	0.7105	-0.010	1.003	0.0171	0.066	0.0022	0.97	0.1635	0.6692								
	9	0.0000	-0.0001	-0.0001	-0.0001	10.4277	-6.5293	0.0160	0.0160	0.0002	0.0002	0.0021	0.0021	0.0879	0.6681								
478	8	1.0047	-0.0001	-0.0001	0.0006	-0.0407	0.7135	-0.011	1.006	0.0389	0.066	0.0049	0.99	0.1268	0.6652								
	8	0.0044	-0.0001	-0.0001	0.0006	-0.0407	0.7135	-0.011	1.006	0.0389	0.066	0.0049	0.99	0.1268	0.6652								
	9	0.0002	-0.0001	-0.0001	-0.0001	4.0465	-2.7523	0.0374	0.0374	0.0006	0.0006	0.0049	0.0049	0.0846	0.6652								
478	9	1.0047	-0.0002	-0.0002	0.0004	-0.1367	0.7276	-0.022	1.10	0.0679	0.066	0.0093	1.02	0.0972	0.6632								
	8	0.0032	-0.0002	-0.0002	0.0004	-0.1367	0.7276	-0.022	1.10	0.0679	0.066	0.0093	1.02	0.0972	0.6632								
	9	0.0011	-0.0002	-0.0002	-0.0002	0.8995	0.1537	0.0707	0.0707	0.0009	0.0009	0.0093	0.0093	0.0656	0.6604								
478	10	1.0046	-0.0001	-0.0001	0.0004	-0.2013	0.6989	-0.022	1.13	0.0980	0.066	0.0124	1.05	0.0863	0.6391								
	8	0.0031	-0.0001	-0.0001	0.0004	-0.2013	0.6989	-0.022	1.13	0.0980	0.066	0.0124	1.05	0.0863	0.6391								
	9	0.0003	-0.0002	-0.0002	-0.0002	3.7860	-0.1140	0.0980	0.0980	0.0011	0.0011	0.0124	0.0124	0.0571	0.6349								
478	11	1.0048	-0.0001	-0.0001	0.0004	-0.2082	0.7278	-0.022	1.16	0.1128	0.066	0.0140	1.07	0.0907	0.6134								
	8	0.0028	-0.0001	-0.0001	0.0004	-0.2082	0.7278	-0.022	1.16	0.1128	0.066	0.0140	1.07	0.0907	0.6134								
	9	-0.0008	-0.0001	-0.0001	-0.0001	-1.0300	0.9665	0.1159	0.1159	0.0015	0.0015	0.0140	0.0140	0.0673	0.6072								
478	12	1.0049	-0.0002	-0.0002	0.0007	-0.0771	0.7089	-0.022	1.00	0.0073	0.066	0.0009	0.96	0.1828	0.6577								
	8	0.0049	-0.0002	-0.0002	0.0007	-0.0771	0.7089	-0.022	1.00	0.0073	0.066	0.0009	0.96	0.1828	0.6577								
	9	-0.0000	-0.0002	-0.0002	-0.0001	-35.1872	14.3295	0.0077	0.0077	0.0000	0.0000	0.0009	0.0009	0.0512	0.6269								
479	1	1.0048	-0.0001	-0.0001	0.0005	-0.1495	0.7151	-0.022	1.11	0.0275	0.066	0.0037	1.06	0.1489	0.6723								
	8	0.0033	-0.0001	-0.0001	0.0005	-0.1495	0.7151	-0.022	1.11	0.0275	0.066	0.0037	1.06	0.1489	0.6723								
	9	-0.0007	-0.0002	-0.0002	-0.0001	-1.5297	1.0291	-0.0233	-0.0233	-0.0007	-0.0007	-0.0032	-0.0032	0.1706	0.6971								
479	2	1.0048	-0.0002	-0.0002	0.0006	-0.1063	0.7313	-0.011	1.07	0.0060	0.066	0.0008	1.03	0.2158	0.6908								
	8	0.0044	-0.0002	-0.0002	0.0006	-0.1063	0.7313	-0.011	1.07	0.0060	0.066	0.0008	1.03	0.2158	0.6908								
	9	-0.0002	-0.0002	-0.0002	-0.0001	-4.9271	2.5024	-0.0045	-0.0045	-0.0003	-0.0003	-0.0006	-0.0006	0.3769	0.7641								

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																						
479	5	1.048	-0.04	-0.00	-0.0745	-0.01	-0.05	0.06	-0.02	0.0002	0.0976	0.5655	0.0047	-0.0002	0.0006	-0.0745	-0.01	-0.05	0.0001	0.0002	0.0021	0.0003	0.0976	0.5655
	6	0.0000	0.0000	0.0000	0.0000	0.7305	0.0021	0.0000	0.0000	0.0000	-0.1782	0.5305	0.0000	0.0000	0.0000	0.0000	0.7305	0.0021	0.0000	0.0000	0.0000	0.0000	-0.1782	0.5305
	9	-0.0000	0.0000	-0.0000	-0.1716	7.1498	0.0028	-0.0028	-0.0001	0.0000							7.1498	0.0028	0.0000	0.0000	0.0000	0.0000		
479	4	1.051	0.49	-0.00	-0.0669	-0.01	-0.05	0.06	-0.03	0.0007	0.1819	0.6341	0.0049	-0.0002	0.0000	-0.0669	-0.01	-0.05	0.0000	0.0000	0.0007	0.0008	0.1819	0.6341
	6	0.0000	0.0000	0.0000	0.0000	0.7202	0.0062	0.0064	0.0000	0.0002	0.0364	0.6462	0.0000	0.0000	0.0000	0.0000	0.7202	0.0062	0.0064	0.0000	0.0000	0.0000	0.0364	0.6462
	9	-0.0000	0.0000	-0.0000	129.1078	210.2695	0.0064	0.0064	-0.0000	0.0000						129.1078	210.2695	0.0064	0.0064	0.0000	0.0000	0.0000		
479	5	1.050	1.01	-0.00	-0.0642	-0.02	-0.04	0.06	-0.01	0.0000	0.1907	0.6754	0.0050	-0.0002	0.0000	-0.0642	-0.02	-0.04	0.0000	0.0000	0.0000	0.0013	0.1907	0.6754
	6	0.0000	0.0000	0.0000	0.0000	0.6864	0.0102	0.0103	0.0000	0.0003	0.0696	0.6377	0.0000	0.0000	0.0000	0.0000	0.6864	0.0102	0.0103	0.0000	0.0003	0.0000	0.0696	0.6377
	9	-0.0000	0.0000	-0.0000	210.2695		0.0103	0.0103	-0.0000	0.0001						210.2695		0.0103	0.0103	0.0000	0.0001	0.0000		
479	6	1.049	3.13	-0.00	-0.0408	-0.02	-0.00	0.06	-0.00	0.0000	0.1354	0.6577	0.0049	-0.0002	0.0000	-0.0408	-0.02	-0.00	0.0000	0.0000	0.0000	0.0041	0.1354	0.6577
	6	0.0000	0.0000	0.0000	10.5971	-4.9580	0.0318	0.0303	0.0000	0.0006	0.0889	0.6575	0.0000	0.0000	0.0000	10.5971	-4.9580	0.0318	0.0303	0.0000	0.0006	0.0000	0.0889	0.6575
	9	0.0000	0.0000	-0.0000	-4.9580		0.0303	0.0303	-0.0000	0.0005						-4.9580		0.0303	0.0303	0.0000	0.0005	0.0000		
479	7	1.049	6.21	0.00	-0.1455	-0.02	0.02	0.06	-0.04	0.0004	0.1013	0.6652	0.0032	-0.0002	0.0000	-0.1455	-0.02	0.02	0.0000	0.0004	0.0012	0.04	0.1013	0.6652
	6	0.0000	0.0000	0.0000	1.0715	0.1860	0.0610	0.0640	0.0000	0.0000	0.0695	0.6597	0.0010	-0.0002	0.0000	1.0715	0.1860	0.0610	0.0640	0.0000	0.0000	0.0084	0.1013	0.6652
	9	0.0010	0.0002	0.0000	-1.1253		0.0640	0.0640	-0.0000	0.0000						-1.1253		0.0640	0.0640	0.0000	0.0000	0.0084		
479	8	1.051	9.35	-0.00	-0.2071	-0.02	0.06	0.06	-0.05	0.0004	0.0859	0.6415	0.0030	-0.0002	0.0000	-0.2071	-0.02	0.06	0.0004	0.0004	0.0015	0.06	0.0859	0.6415
	6	0.0000	0.0000	0.0000	5.3114	-1.1253	0.0922	0.0922	0.0000	0.0000	0.0570	0.6397	0.0002	-0.0002	0.0000	5.3114	-1.1253	0.0922	0.0922	0.0000	0.0000	0.0113	0.0570	0.6397
	9	0.0002	0.0002	-0.0000	-1.1253		0.0922	0.0922	-0.0000	0.0000						-1.1253		0.0922	0.0922	0.0000	0.0000	0.0113		
479	9	1.051	12.57	0.00	-0.2011	-0.02	0.09	0.06	-0.05	0.0004	0.0851	0.6214	0.0004	-0.0001	0.0000	-0.2011	-0.02	0.09	0.0004	0.0004	0.0018	0.08	0.0851	0.6214
	6	0.0000	0.0000	0.0000	-1.1382	1.0401	0.1107	0.1129	0.0000	0.0000	0.0660	0.6135	0.0008	-0.0001	0.0000	-1.1382	1.0401	0.1107	0.1129	0.0000	0.0000	0.0137	0.0660	0.6135
	9	0.0008	0.0001	-0.0001	1.0401		0.1129	0.1129	-0.0000	0.0000						1.0401		0.1129	0.1129	0.0000	0.0000	0.0137		
479	10	1.047	-0.04	-0.00	-0.0635	-0.01	-0.05	0.06	-0.05	0.0007	0.0410	0.4609	0.0045	-0.0002	0.0000	-0.0635	-0.01	-0.05	0.0007	0.0000	0.0000	0.0002	0.0410	0.4609
	6	0.0000	0.0000	0.0000	-4.0134	1.3935	0.0030	0.0030	0.0000	0.0000	0.1776	0.4490	0.0003	-0.0000	0.0000	-4.0134	1.3935	0.0030	0.0030	0.0000	0.0000	0.0002	0.1776	0.4490
	9	0.0003	0.0002	-0.0000	1.3935		0.0030	0.0030	-0.0000	0.0000						1.3935		0.0030	0.0030	0.0000	0.0000	0.0002		
480	1	1.048	-3.11	-0.00	-0.1553	-0.02	-0.05	0.06	-0.05	0.0005	0.1374	0.6739	0.0033	-0.0002	0.0000	-0.1553	-0.02	-0.05	0.0005	0.0000	0.0009	0.0045	0.1374	0.6739
	6	0.0000	0.0000	0.0000	-1.3544	1.0550	0.0337	0.0302	0.0000	0.0000	0.1455	0.6916	0.0008	-0.0002	0.0000	-1.3544	1.0550	0.0337	0.0302	0.0000	0.0000	0.0041	0.1455	0.6916
	9	0.0008	0.0002	-0.0001	1.0550		0.0302	0.0302	-0.0000	0.0000						1.0550		0.0302	0.0302	0.0000	0.0000	0.0041		
480	2	1.051	-1.06	-0.00	-0.1178	-0.01	-0.09	0.06	-0.09	0.0006	0.1889	0.6854	0.0044	-0.0002	0.0000	-0.1178	-0.01	-0.09	0.0006	0.0000	0.0004	0.0015	0.1889	0.6854
	6	0.0000	0.0000	0.0000	-5.3256	3.3146	0.0097	0.0097	0.0000	0.0000	0.2575	0.7418	0.0002	-0.0001	0.0000	-5.3256	3.3146	0.0097	0.0097	0.0000	0.0000	0.0014	0.2575	0.7418
	9	0.0002	0.0002	-0.0001	3.3146		0.0097	0.0097	-0.0000	0.0000						3.3146		0.0097	0.0097	0.0000	0.0000	0.0014		
480	3	1.051	-0.03	-0.00	-0.0736	-0.01	-0.09	0.06	-0.09	0.0007	0.3659	0.9352	0.0049	-0.0002	0.0000	-0.0736	-0.01	-0.09	0.0007	0.0000	0.0001	0.0003	0.3659	0.9352
	6	0.0000	0.0000	0.0000	-8.3787	4.0355	0.0020	0.0024	0.0000	0.0000	0.5273	0.9176	0.0001	-0.0002	0.0000	-8.3787	4.0355	0.0020	0.0024	0.0000	0.0000	0.0003	0.5273	0.9176
	9	0.0001	0.0002	-0.0001	4.0355		0.0024	0.0024	-0.0000	0.0000						4.0355		0.0024	0.0024	0.0000	0.0000	0.0003		

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key											
480	4	7	1.049	0.48	-0.007	-0.0595	-0.01	-0.95	0.06	-1.00	0.0921	0.4032	
		8	0.0046	-0.0000	0.0000	0.7712	0.0017	0.0000	0.0001	0.0001	0.3951	0.3004	
		9	-0.0000	0.0002	-0.0001	31.3712	0.0016	-0.0001	0.0001	0.0001	-0.0001	0.0001	
480	5	7	1.048	1.02	-0.007	-0.0635	-0.01	-0.94	0.06	-1.00	0.2027	0.6276	
		8	0.0050	-0.0000	0.0000	0.7103	0.0056	0.0002	0.0000	0.0000	0.0035	0.5318	
		9	-0.0000	0.0002	-0.0001	13.1880	0.0051	-0.0000	0.0005	0.0005	-0.0005	0.0005	
480	6	7	1.051	3.08	-0.006	-0.0480	-0.01	-0.91	0.06	-0.97	0.1468	0.6479	
		8	0.0045	-0.0000	0.0000	0.7104	0.0253	0.0007	0.0004	0.0032	0.0958	0.6386	
		9	-0.0002	0.0002	-0.0001	2.5024	0.0232	0.0004	0.0004	0.0029	0.0004	0.0004	
480	7	7	1.048	6.23	0.004	-0.1673	-0.02	-0.87	0.06	-0.95	0.1074	0.6659	
		8	0.0032	-0.0001	0.0000	0.6930	0.0552	0.0011	0.0008	0.0075	0.0729	0.6522	
		9	0.0008	0.0002	0.0000	0.0118	0.0575	0.0000	0.0000	0.0000	0.0000	0.0000	
480	8	7	1.047	9.36	-0.004	-0.1997	-0.02	-0.83	0.06	-0.91	0.0878	0.6469	
		8	0.0028	-0.0001	0.0000	0.7708	0.0842	0.0014	0.0014	0.0108	0.0579	0.6471	
		9	0.0000	0.0002	-0.0000	0.0406	0.0874	0.0010	0.0010	0.0113	0.0000	0.0000	
480	9	7	1.050	12.56	-0.002	-0.2310	-0.02	-0.80	0.05	-0.88	0.0819	0.6257	
		8	0.0027	-0.0001	0.0004	0.7561	0.1080	0.0017	0.0017	0.0135	0.0587	0.6195	
		9	-0.0010	0.0002	-0.0002	1.1181	0.1108	0.0013	0.0013	0.0137	0.0000	0.0000	
480	10	7	1.049	-0.05	-0.006	-0.0780	-0.01	-0.96	0.06	-1.01	0.4103	0.9873	
		8	0.0046	-0.0000	0.0000	0.7503	0.0021	-0.0001	-0.0001	0.0004	0.5531	0.9668	
		9	-0.0002	0.0002	-0.0001	2.0668	-0.0024	-0.0002	-0.0002	0.0004	0.0000	0.0000	
481	1	7	1.046	-3.12	-0.004	-0.1732	-0.02	-3.00	0.06	-3.01	0.1176	0.6828	
		8	0.0034	-0.0001	0.0004	0.7186	-0.0471	-0.0011	-0.0011	0.0064	0.1147	0.6884	
		9	-0.0011	0.0002	-0.0001	0.8338	-0.0433	-0.0009	-0.0009	0.0059	0.0000	0.0000	
481	2	7	1.048	-1.09	-0.006	-1.142	-0.01	-2.98	0.06	-2.98	0.1415	0.6828	
		8	0.0042	-0.0000	0.0000	0.7179	-0.0254	-0.0007	-0.0007	0.0034	0.1530	0.6992	
		9	-0.0006	0.0002	-0.0001	1.1363	-0.0227	-0.0006	-0.0006	0.0031	0.0000	0.0000	
481	3	7	1.051	-0.06	-0.006	-0.0709	-0.01	-2.95	0.06	-2.98	0.1579	0.6866	
		8	0.0045	-0.0000	0.0006	0.7625	-0.0142	-0.0004	-0.0004	0.0019	0.2073	0.7307	
		9	-0.0003	0.0002	-0.0001	1.9970	-0.0126	-0.0005	-0.0005	0.0018	0.0000	0.0000	
481	4	7	1.051	0.47	-0.006	-0.0504	-0.01	-2.95	0.06	-2.96	0.1882	0.7287	
		8	0.0049	-0.0000	0.0007	0.7158	-0.0090	-0.0003	-0.0003	0.0013	0.2487	0.7596	
		9	-0.0000	0.0002	-0.0001	8.3102	-0.0086	-0.0004	-0.0004	0.0013	0.0000	0.0000	

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key															
481	5	7	1.051	0.99	-0.00	-0.0575	-0.01	-2.94	0.06	-2.97	0.1932	0.7772						
		8	0.0051	-0.0000	0.0007	37.7760	0.6874	-0.0042	-0.0001	-0.0006	0.2928	0.7697						
		9	0.0000	0.0002	-0.0001	-0.0001	-0.3326	-0.0049	-0.0002	-0.0007	0.0000	0.0000						
481	6	7	1.052	3.06	-0.00	-0.0487	-0.02	-2.91	0.06	-2.94	0.1877	0.6220						
		8	0.0045	-0.0000	0.0006	-9.7479	0.6926	0.0124	0.0004	0.0015	0.1076	0.6306						
		9	-0.0001	0.0002	-0.0001	-0.0001	4.9883	0.0109	0.0002	0.0013	0.0000	0.0000						
481	7	7	1.047	6.21	0.00	-0.1714	-0.02	-2.87	0.06	-2.90	0.1290	0.6567						
		8	0.0033	-0.0001	0.0004	1.2635	0.6293	0.0420	0.0010	0.0055	0.0902	0.6530						
		9	0.0000	0.0002	-0.0000	-0.0000	-0.1251	0.0440	0.0007	0.0057	0.0000	0.0000						
481	8	7	1.052	9.35	0.00	-0.2074	-0.02	-2.83	0.06	-2.87	0.0941	0.6535						
		8	0.0030	-0.0001	0.0004	3.8912	0.6625	0.0724	0.0013	0.0094	0.0645	0.6512						
		9	0.0002	0.0002	-0.0000	-0.0000	-0.4656	0.0753	0.0009	0.0098	0.0000	0.0000						
481	9	7	1.049	12.54	-0.00	-0.2335	-0.02	-2.80	0.05	-2.85	0.0805	0.6338						
		8	0.0028	-0.0001	0.0003	-0.9471	0.6930	0.0996	0.0016	0.0126	0.0546	0.6338						
		9	-0.0011	0.0002	-0.0002	-0.0002	0.9922	0.1030	0.0011	0.0130	0.0000	0.0000						
481	10	7	1.052	-0.07	-0.00	-0.0677	-0.01	-2.96	0.06	-2.98	0.1606	0.6984						
		8	0.0046	-0.0000	0.0006	-3.2079	0.7450	-0.0143	-0.0004	-0.0020	0.2121	0.7527						
		9	-0.0004	0.0002	-0.0001	-0.0001	1.4904	-0.0127	-0.0005	-0.0019	0.0000	0.0000						
482	7	7	0.900	-3.11	-0.00	-0.1112	-0.02	-3.01	0.05	-3.00	0.1533	0.6641						
		8	0.0040	-0.0000	0.0005	-0.5625	0.7372	-0.0494	-0.0015	-0.0065	0.1343	0.6673						
		9	-0.0014	0.0001	-0.0002	-0.0002	0.8966	-0.0474	-0.0012	-0.0063	0.0000	0.0000						
482	8	7	0.900	-1.07	-0.00	-0.0268	-0.02	-2.97	0.06	-2.98	0.1358	0.7059						
		8	0.0047	-0.0000	0.0006	-1.0823	0.6509	-0.0276	-0.0007	-0.0039	0.1377	0.6858						
		9	-0.0008	0.0001	-0.0001	-0.0001	0.8796	-0.0255	-0.0007	-0.0035	0.0000	0.0000						
482	9	7	0.900	-0.06	-0.00	0.0268	-0.02	-2.95	0.06	-2.96	0.1552	0.7486						
		8	0.0047	0.0000	0.0006	-1.7693	0.6665	-0.0161	-0.0005	-0.0024	0.1564	0.7004						
		9	-0.0005	0.0001	-0.0001	-0.0001	1.3745	-0.0152	-0.0004	-0.0021	0.0000	0.0000						
482	10	7	0.899	0.48	-0.00	0.0564	-0.02	-2.94	0.06	-2.96	0.2073	0.7980						
		8	0.0048	0.0000	0.0006	-1.4797	0.6737	-0.0098	-0.0004	-0.0015	0.1963	0.7372						
		9	-0.0007	0.0002	-0.0001	-0.0001	0.8327	-0.0098	-0.0003	-0.0014	0.0000	0.0000						
482	11	7	0.900	0.98	-0.00	0.0823	-0.02	-2.93	0.06	-2.95	0.3277	0.9607						
		8	0.0050	0.0000	0.0006	-1.4509	0.6429	-0.0043	-0.0002	-0.0008	0.2796	0.8459						
		9	-0.0007	0.0002	-0.0001	-0.0001	0.9413	-0.0053	-0.0002	-0.0009	0.0000	0.0000						

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																					
482	12	7	0.900	5.07	-0.000	0.0934	-0.02	-2.91	0.06	-2.93	0.1130	0.6673	8	0.0042	0.0000	0.0005	0.6572	0.0141	0.0003	0.0018	0.0762	0.6723	
		9	-0.0008	0.0000	-0.0002	-1.5468	1.2318	0.0127	0.0001	0.0017													
482	13	7	0.901	6.16	-0.000	0.0698	-0.02	-2.87	0.06	-2.90	0.1353	0.6571	8	0.0035	-0.0004	0.0004	0.6510	0.0457	0.0012	0.0060	0.1116	0.6551	
		9	-0.0001	0.0003	-0.0001	-9.0055	3.9687	0.0479	0.0010	0.0062													
482	14	7	0.901	9.28	-0.000	0.1396	-0.02	-2.83	0.06	-2.87	0.1332	0.6192	8	0.0036	-0.0004	0.0004	0.6306	0.0759	0.0020	0.0093	0.1050	0.6200	
		9	-0.0001	0.0002	-0.0001	-7.2141	3.2251	0.0787	0.0016	0.0097													
482	15	7	0.900	12.44	-0.000	0.1554	-0.02	-2.80	0.06	-2.85	0.1194	0.5998	8	0.0032	-0.0004	0.0004	0.6906	0.0997	0.0023	0.0119	0.0992	0.5981	
		9	-0.0012	0.0002	-0.0002	-0.8266	0.8641	0.1015	0.0020	0.0121													
482	16	7	0.900	-0.07	-0.000	0.0264	-0.02	-2.95	0.06	-2.97	0.1582	0.7587	8	0.0049	0.0006	0.0006	0.6549	0.0160	0.0005	0.0024	0.1538	0.7256	
		9	-0.0010	0.0002	-0.0001	-1.2414	0.5945	-0.0155	0.0004	0.0022													
483	1	7	0.901	-3.10	-0.000	0.1268	-0.02	-1.00	0.06	-1.04	0.1447	0.6886	8	0.0040	-0.0001	0.0005	0.6858	0.0367	0.0010	0.0050	0.1360	0.6702	
		9	-0.0016	0.0001	-0.0002	-0.5959	0.7834	-0.0347	0.0009	0.0046													
483	2	7	0.901	-1.08	-0.000	0.0059	-0.02	-0.97	0.06	-1.02	0.1783	0.7385	8	0.0047	-0.0001	0.0006	0.6760	0.0131	0.0004	0.0019	0.1674	0.6943	
		9	-0.0006	0.0001	-0.0002	-1.4660	1.5870	-0.0127	0.0004	0.0017													
483	3	7	0.901	-0.05	-0.000	0.0417	-0.02	-0.95	0.06	-1.00	0.4699	1.0418	8	0.0049	0.0000	0.0006	0.6591	0.0022	0.0006	0.0004	0.5936	0.8348	
		9	-0.0007	0.0002	-0.0001	-1.4573	1.1908	-0.0025	0.0003	0.0004													
483	4	7	0.900	0.49	-0.000	0.0769	-0.01	-0.95	0.06	-1.00	0.0205	0.5077	8	0.0051	0.0000	0.0006	0.6505	0.0023	0.0000	0.0000	-0.6182	0.7309	
		9	-0.0006	0.0002	-0.0001	-1.7933	1.5621	0.0011	0.0001	0.0001													
483	5	7	0.901	1.01	-0.000	0.0943	-0.01	-0.94	0.06	-1.00	0.2414	0.6547	8	0.0051	0.0000	0.0006	0.6554	0.0056	0.0002	0.0005	0.0729	0.6606	
		9	-0.0005	0.0002	-0.0001	-2.3036	1.8024	0.0044	0.0000	0.0005													
483	6	7	0.901	3.10	-0.000	0.0629	-0.02	-0.90	0.06	-0.97	0.1262	0.6689	8	0.0045	0.0000	0.0005	0.6476	0.0283	0.0007	0.0035	0.1877	0.6686	
		9	-0.0010	0.0003	-0.0001	-1.4238	0.9156	0.0267	0.0004	0.0004													

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																		
483	7	0.903	6.21	-0.00	-0.02	-0.85	0.06	-0.93	0.1372	0.6453	0.0036	-0.0000	0.0004	-0.1022	0.6284	0.0588	0.0016	0.0075	0.1041	0.6512
	6	-0.0005	0.0003	-0.0001	1.0920	0.0621	0.0012	0.0080												
	5																			
483	7	0.901	9.30	-0.00	-0.02	-0.82	0.06	-0.91	0.1354	0.6070	0.0035	-0.0001	0.0004	-0.1602	0.5863	0.0848	0.0022	0.0103	0.1039	0.6039
	6	-0.0008	0.0002	-0.0001	0.6791	0.0879	0.0018	0.0106												
	5																			
483	7	0.901	12.45	-0.00	-0.02	-0.80	0.06	-0.89	0.1176	0.5905	0.0031	-0.0001	0.0003	-0.1821	0.6250	0.1066	0.0025	0.0125	0.0956	0.5838
	6	-0.0016	0.0002	-0.0002	0.7206	0.1080	0.0020	0.0126												
	5																			
483	7	0.901	-0.05	-0.00	-0.02	-0.95	0.06	-1.00	0.6788	1.3314	0.0050	0.0000	0.0006	0.0445	0.6505	-0.0018	0.0002	-1.00	0.6005	1.0047
	6	-0.0012	0.0002	-0.0001	0.5561	-0.0024	-0.0002	-0.0004												
	5																			
484	7	0.901	-3.09	-0.00	-0.02	-0.11	0.06	-0.05	0.1481	0.6921	0.0042	-0.0000	0.0005	-0.1144	0.6738	-0.0300	0.0008	-0.0041	0.1365	0.6747
	6	-0.0015	0.0002	-0.0002	0.8119	-0.0279	-0.0007	-0.0037												
	5																			
484	7	0.902	-1.08	-0.00	-0.02	-0.08	0.06	-0.03	0.2709	0.7674	0.0047	0.0000	0.0006	0.0088	0.6508	-0.0066	0.0003	-0.0010	0.2444	0.8120
	6	-0.0008	0.0002	-0.0001	1.1041	-0.0062	-0.0003	-0.0009												
	5																			
484	7	0.900	-0.05	-0.00	-0.01	-0.07	0.06	-0.02	0.0267	0.6925	0.0050	0.0000	0.0006	0.0743	0.6589	0.0026	0.0000	0.0002	0.0267	0.5577
	6	-0.0008	0.0002	-0.0001	1.0508	0.0021	-0.0000	0.0003												
	5																			
484	7	0.901	0.49	-0.00	-0.01	-0.05	0.06	-0.03	0.2627	0.7119	0.0051	0.0000	0.0006	0.0789	0.6330	0.0059	0.0003	0.0008	0.1085	0.7058
	6	-0.0007	0.0002	-0.0001	1.0171	0.0058	0.0001	0.0008												
	5																			
484	7	0.901	0.96	-0.00	-0.01	-0.04	0.06	-0.01	0.1401	0.7175	0.0050	0.0000	0.0006	0.0973	0.6565	0.0116	0.0003	0.0016	0.0743	0.7175
	6	-0.0007	0.0002	-0.0002	1.3333	0.0110	0.0001	0.0015												
	5																			
484	7	0.901	3.10	-0.00	-0.02	-0.02	0.06	0.00	0.1223	0.6645	0.0045	0.0000	0.0005	0.0493	0.6334	0.0350	0.0008	0.0046	0.0914	0.6645
	6	-0.0010	0.0002	-0.0002	1.0049	0.0337	0.0006	0.0045												
	5																			
484	7	0.901	6.21	-0.00	-0.02	-0.02	0.06	0.03	0.1364	0.6338	0.0037	0.0000	0.0004	0.1104	0.6120	0.0650	0.0017	0.0082	0.1047	0.6338
	6	-0.0005	0.0003	-0.0001	1.1787	0.0681	0.0014	0.0086												
	5																			

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																		
484	8	0.500	0.0035	-0.0007	0.0000	0.0003	-0.0001	0.0004	-0.1367	0.6194	0.0886	0.0912	0.05	0.06	0.0023	0.0106	0.0109	0.1344	0.6023	0.5985
484	9	0.899	0.0031	-0.0015	12.45	0.0001	0.0003	-0.0002	-0.1898	0.6234	0.1109	0.1126	0.08	0.06	0.0025	0.0130	0.0131	0.1133	0.5885	0.5834
484	10	0.899	0.0052	-0.0011	-0.04	0.0003	0.0006	-0.0001	0.0386	0.6040	0.0029	0.0025	-0.06	0.06	0.0000	0.0002	0.0002	-0.0327	0.4584	0.5366
485	1	0.899	0.0041	-0.0016	-3.07	0.0002	0.0005	-0.0002	-0.1115	0.6652	0.0221	0.0207	0.96	0.06	0.0007	0.0031	0.0028	0.1588	0.7029	0.6849
485	2	0.899	0.0047	-0.0010	-1.05	0.0002	0.0006	-0.0002	0.0106	0.6620	0.0008	0.0006	0.99	0.06	0.0001	0.0000	0.0000	-0.9362	0.0782	0.7798
485	3	0.901	0.0048	-0.0008	-0.03	0.0002	0.0006	-0.0001	0.0555	0.6543	0.0075	0.0075	1.01	0.05	0.0001	0.0010	0.0010	0.2061	0.6884	0.7143
485	4	0.900	0.0050	-0.0008	0.49	0.0002	0.0006	-0.0002	0.0827	0.6523	0.0137	0.0133	1.01	0.06	0.0001	0.0019	0.0018	0.1122	0.7001	0.6949
485	5	0.899	0.0051	-0.0009	0.99	0.0002	0.0006	-0.0002	0.0794	0.6421	0.0203	0.0188	1.03	0.06	0.0002	0.0025	0.0025	0.0991	0.6723	0.6709
485	6	0.900	0.0046	-0.0011	3.06	0.0003	0.0005	-0.0002	0.0429	0.6216	0.0430	0.0411	1.06	0.06	0.0010	0.0056	0.0055	0.1249	0.6576	0.6699
485	7	0.900	0.0037	-0.0008	6.23	0.0003	0.0004	-0.0001	-0.1232	0.5713	0.0703	0.0722	1.10	0.06	0.0019	0.0089	0.0089	0.1400	0.6180	0.6208
485	8	0.899	0.0031	-0.0006	9.33	0.0002	0.0004	-0.0001	-0.1453	0.6805	0.0925	0.0540	1.13	0.06	0.0024	0.0109	0.0111	0.1314	0.5929	0.5939

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key													
485	9	7	0.899	12.49	-0.000	-0.000	-0.000	-0.1874	-0.02	1.15	0.06	1.06	0.1117	0.5837		
		8	0.0030	-0.0001	0.0003	0.0003	-0.1874	0.6257	0.1130	0.0025	0.0131	0.0913	0.5825			
		9	-0.0015	0.0002	-0.0002	-0.0002	-0.7184	0.9072	0.1148	0.0020	0.0133					
485	10	7	0.901	-0.03	-0.000	-0.000	0.0495	-0.01	1.01	0.06	0.95	0.1709	0.6465			
		8	0.0049	0.0000	0.0006	0.0006	0.0495	0.6429	0.0079	0.0002	0.0010	0.1058	0.6749			
		9	-0.0012	0.0003	-0.0001	-0.0001	-1.2152	0.6786	0.0076	0.0001	0.0010					
486	1	7	0.898	-3.07	-0.000	-0.000	-0.0927	-0.02	2.95	0.06	3.03	0.2699	0.7790			
		8	0.0042	-0.0000	0.0005	-0.0005	-0.0927	0.6702	-0.0072	-0.0003	-0.0011	0.2625	0.7528			
		9	-0.0013	0.0002	-0.0002	-0.0002	-0.8251	0.9275	-0.0056	-0.0002	-0.0008					
486	2	7	0.899	-1.05	-0.000	-0.000	0.0183	-0.02	2.98	0.07	3.05	0.1351	0.7256			
		8	0.0046	0.0000	0.0005	0.0005	0.0183	0.6345	0.0108	0.0002	0.0015	0.0667	0.7135			
		9	-0.0010	0.0002	-0.0001	-0.0001	-1.2927	0.9378	0.0128	0.0001	0.0018					
486	3	7	0.900	-0.04	-0.000	-0.000	0.0617	-0.01	2.99	0.06	3.07	0.1034	0.6761			
		8	0.0047	0.0000	0.0006	0.0006	0.0617	0.6662	0.0230	0.0004	0.0031	0.0833	0.6768			
		9	-0.0009	0.0002	-0.0002	-0.0002	-1.5198	1.2069	0.0236	0.0003	0.0032					
486	4	7	0.899	0.51	-0.000	-0.000	0.0678	-0.01	3.00	0.07	3.07	0.1146	0.6875			
		8	0.0051	0.0000	0.0006	0.0006	0.0678	0.6388	0.0282	0.0006	0.0038	0.0832	0.6768			
		9	-0.0010	0.0002	-0.0001	-0.0001	-1.4619	0.9582	0.0287	0.0004	0.0038					
486	5	7	0.901	1.03	-0.000	-0.000	0.0615	-0.01	3.01	0.06	3.08	0.1159	0.6760			
		8	0.0052	0.0000	0.0006	0.0006	0.0615	0.6193	0.0343	0.0007	0.0046	0.0892	0.6766			
		9	-0.0010	0.0002	-0.0002	-0.0002	-1.4124	1.2025	0.0341	0.0006	0.0046					
486	6	7	0.899	3.13	-0.000	-0.000	0.0135	-0.02	3.04	0.06	3.11	0.1380	0.6460			
		8	0.0047	0.0000	0.0006	0.0006	0.0135	0.6428	0.0555	0.0015	0.0073	0.1061	0.6564			
		9	-0.0012	0.0003	-0.0002	-0.0002	-1.3176	1.0212	0.0560	0.0011	0.0073					
486	7	7	0.900	6.20	-0.000	-0.000	-0.1112	-0.02	3.08	0.06	3.14	0.1466	0.6025			
		8	0.0037	-0.0000	0.0004	0.0004	-0.1112	0.5872	0.0784	0.0023	0.0094	0.1200	0.6028			
		9	-0.0006	0.0003	-0.0001	-0.0001	-2.9180	1.3697	0.0812	0.0019	0.0097					
486	8	7	0.900	9.34	-0.000	-0.000	-0.1563	-0.02	3.10	0.07	3.16	0.1281	0.5897			
		8	0.0032	-0.0001	0.0004	0.0004	-0.1563	0.6212	0.1002	0.0025	0.0118	0.1043	0.5824			
		9	-0.0007	0.0002	-0.0002	-0.0002	-1.8987	1.3868	0.1022	0.0021	0.0119					
486	9	7	0.899	12.47	-0.000	-0.000	-0.1792	-0.02	3.11	0.06	3.17	0.1017	0.5807			
		8	0.0031	-0.0001	0.0003	0.0003	-0.1792	0.6424	0.1162	0.0023	0.0135	0.0807	0.5778			
		9	-0.0015	0.0002	-0.0002	-0.0002	-0.7914	0.8305	0.1202	0.0019	0.0139					

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Rur.	Pt.	Cd	See Key											
486	10	7 6 9	0.900 0.0049 -0.0011	-0.000 0.0003 0.0003	-0.00 0.0006 -0.0002	0.0619 -1.3426 0.0000	-0.01 0.6559 0.9074	2.229 0.0232 0.0000	0.06 0.0004 0.0003	3.08 0.0030 0.0031	0.1004 0.0834 0.0000	0.6713 0.6729 0.0000		
487	1	7 6 9	0.899 0.0028 -0.0012	-3.03 0.0000 0.0001	-0.00 0.0005 -0.0002	-0.0825 -0.7761 0.0000	-0.02 0.6684 1.0997	6.01 0.0108 0.0126	0.06 0.0002 0.0001	6.00 0.0015 0.0018	0.1119 0.0781 0.0000	0.7274 0.7418 0.0000		
487	2	7 6 9	0.898 0.0046 -0.0010	-1.01 0.0000 0.0002	-0.00 0.0005 -0.0002	0.0170 -1.2723 0.0000	-0.02 0.6042 1.0378	6.04 0.333 0.0353	0.06 0.0007 0.0006	6.02 0.0045 0.0048	0.1087 0.0859 0.0000	0.6872 0.6860 0.0000		
487	3	7 6 9	0.899 0.0051 -0.0009	-0.00 0.0000 0.0002	-0.00 0.0006 -0.0002	0.0269 -1.5031 0.0000	-0.02 0.6270 1.2581	6.06 0.0439 0.0449	0.06 0.0010 0.0008	6.03 0.0059 0.0060	0.1242 0.0982 0.0000	0.6736 0.6746 0.0000		
487	4	7 6 9	0.900 0.0055 -0.0011	0.52 0.0000 0.0003	-0.00 0.0006 -0.0002	0.0216 -1.3563 0.0000	-0.02 0.6086 1.0285	6.07 0.0490 0.0497	0.07 0.0012 0.0010	6.04 0.0065 0.0066	0.1317 0.1041 0.0000	0.6626 0.6688 0.0000		
487	5	7 6 9	0.899 0.0055 -0.0012	1.05 0.0000 0.0003	-0.00 0.0007 -0.0002	0.0391 -1.2617 0.0000	-0.01 0.6305 0.9444	6.07 0.0543 0.0548	0.07 0.0014 0.0011	6.05 0.0070 0.0072	0.1359 0.1039 0.0000	0.6456 0.6584 0.0000		
487	6	7 6 9	0.899 0.0048 -0.0013	3.13 0.0000 0.0003	-0.00 0.0006 -0.0002	-0.0008 -1.2292 0.0000	-0.01 0.6190 0.9635	6.10 0.0704 0.0713	0.06 0.0021 0.0017	6.07 0.0086 0.0088	0.1515 0.1200 0.0000	0.6111 0.6194 0.0000		
487	7	7 6 9	0.900 0.0034 -0.0007	6.25 0.0000 0.0003	-0.00 0.0004 -0.0001	-0.1269 -2.1308 0.0000	-0.02 0.6228 1.2455	6.13 0.0910 0.0928	0.06 0.0025 0.0021	6.10 0.0107 0.0109	0.1378 0.1178 0.0000	0.5925 0.5871 0.0000		
487	8	7 6 9	0.899 0.0034 -0.0008	9.32 0.0001 0.0002	-0.00 0.0004 -0.0002	-0.1666 -1.5462 0.0000	-0.02 0.5980 1.1256	6.14 0.1072 0.1104	0.07 0.0025 0.0020	6.10 0.0124 0.0127	0.1179 0.0947 0.0000	0.5782 0.5752 0.0000		
487	9	7 6 9	0.896 0.0033 -0.0013	12.48 0.0001 0.0002	0.00 0.0004 -0.0002	-0.1849 -0.7825 0.0000	-0.02 0.6085 1.0591	6.14 0.1204 0.1205	0.06 0.0020 0.0015	6.10 0.0139 0.0138	0.0857 0.0662 0.0000	0.5793 0.5751 0.0000		
487	10	7 6 9	0.898 0.0053 -0.0013	-0.00 0.0000 0.0003	-0.00 0.0006 -0.0002	0.0278 -1.3198 0.0000	-0.01 0.6097 0.8558	6.06 0.0441 0.0451	0.06 0.0010 0.0008	6.03 0.0058 0.0060	0.1235 0.0983 0.0000	0.6645 0.6707 0.0000		

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

← See Key →

488	1	7	0.898 0.0040 -0.0014	-0.0002 -0.0001 0.0002	-0.0004 0.0000 -0.0002	-0.0002 0.0000 -0.0002	-0.0002 0.0000 0.0000	-0.0002 0.0000 0.0000	-0.0002 0.0000 0.0000	0.0004 -0.0735 -0.8255	-0.001 0.5981 0.9092	9.07 0.0324 0.0342	0.0006 0.0006 0.0006	0.0045 0.0048 0.0048	0.1074 0.0892 0.0892	0.6989 0.7033 0.7033
488	2	7	0.898 0.0047 -0.0011	-0.0002 0.0000 0.0002	-0.0006 0.0000 -0.0002	-0.0002 0.0000 -0.0002	-0.0002 0.0000 -0.0002	-0.0002 0.0000 -0.0002	-0.0002 0.0000 -0.0002	0.0184 -1.2490 -1.2490	-0.001 0.6735 1.1524	9.10 0.0526 0.0548	0.0014 0.0011 0.0011	0.0068 0.0071 0.0071	0.1382 0.1044 0.1044	0.6554 0.6554 0.6554
488	3	7	0.898 0.0054 -0.0011	0.0000 0.0003 0.0003	-0.0006 0.0000 -0.0002	-0.0002 0.0000 -0.0002	-0.0002 0.0000 -0.0002	-0.0002 0.0000 -0.0002	0.0080 -1.3594 -1.3594	0.6131 1.2026 1.2026	9.11 0.0605 0.0624	0.0018 0.0014 0.0014	0.0076 0.0079 0.0079	0.0076 0.0079 0.0079	0.1522 0.1134 0.1134	0.6331 0.6331 0.6331
488	4	7	0.898 0.0056 -0.0010	0.0000 0.0003 0.0003	-0.0006 0.0000 -0.0002	-0.0002 0.0000 -0.0002	-0.0002 0.0000 -0.0002	-0.0002 0.0000 -0.0002	0.0028 -1.4256 -1.4256	0.6119 1.2976 1.2976	9.12 0.0647 0.0661	0.0019 0.0015 0.0015	0.0081 0.0083 0.0083	0.0081 0.0083 0.0083	0.1536 0.1172 0.1172	0.6281 0.6330 0.6330
488	5	7	0.899 0.0057 -0.0012	-0.0003 0.0000 0.0003	-0.0007 0.0000 -0.0002	-0.0002 0.0000 -0.0002	-0.0002 0.0000 -0.0002	-0.0002 0.0000 -0.0002	-0.0012 -1.2455 -1.2455	0.6108 1.0288 1.0288	9.13 0.0684 0.0701	0.0021 0.0016 0.0016	0.0084 0.0086 0.0086	0.0084 0.0086 0.0086	0.1568 0.1177 0.1177	0.6171 0.6183 0.6183
488	6	7	0.897 0.0051 -0.0010	0.0000 0.0003 0.0003	-0.0006 0.0000 -0.0002	-0.0002 0.0000 -0.0002	-0.0002 0.0000 -0.0002	-0.0002 0.0000 -0.0002	-0.0132 -1.5134 -1.5134	0.6427 1.2589 1.2589	9.16 0.0841 0.0844	0.0024 0.0021 0.0021	0.0100 0.0101 0.0101	0.0100 0.0101 0.0101	0.1476 0.1250 0.1250	0.5973 0.5984 0.5984
488	7	7	0.896 0.0037 -0.0004	0.0000 0.0003 0.0003	-0.0004 0.0000 -0.0001	-0.0001 0.0000 -0.0001	-0.0001 0.0000 -0.0001	-0.0001 0.0000 -0.0001	-0.1100 -3.8865 -3.8865	0.6244 1.3948 1.3948	9.17 0.1026 0.1039	0.0025 0.0021 0.0021	0.0119 0.0120 0.0120	0.0119 0.0120 0.0120	0.1264 0.1049 0.1049	0.5816 0.5780 0.5780
488	8	7	0.897 0.0037 -0.0007	0.0000 0.0003 0.0003	-0.0004 0.0000 -0.0001	-0.0001 0.0000 -0.0001	-0.0001 0.0000 -0.0001	-0.0001 0.0000 -0.0001	-0.1288 -2.2165 -2.2165	0.5821 1.0359 1.0359	9.17 0.1128 0.1150	0.0021 0.0017 0.0017	0.0130 0.0131 0.0131	0.0130 0.0131 0.0131	0.0962 0.0746 0.0746	0.5764 0.5718 0.5718
488	9	7	0.897 0.0031 -0.0013	0.0000 0.0003 0.0003	-0.0004 0.0000 -0.0002	-0.0001 0.0000 -0.0001	-0.0001 0.0000 -0.0001	-0.0001 0.0000 -0.0001	-0.1530 -0.9421 -0.9421	0.6444 0.9498 0.9498	9.17 0.1193 0.1222	0.0019 0.0015 0.0015	0.0137 0.0138 0.0138	0.0137 0.0138 0.0138	0.0820 0.0639 0.0639	0.5739 0.5684 0.5684
488	10	7	0.898 0.0055 -0.0014	0.0000 0.0003 0.0003	-0.0006 0.0000 -0.0002	-0.0002 0.0000 -0.0002	-0.0002 0.0000 -0.0002	-0.0002 0.0000 -0.0002	0.0041 -1.2499 -1.2499	0.5948 0.8917 0.8917	9.11 0.0608 0.0626	0.0018 0.0014 0.0014	0.0076 0.0079 0.0079	0.0076 0.0079 0.0079	0.1488 0.1119 0.1119	0.6307 0.6327 0.6327
489	1	7	0.896 0.0038 -0.0013	-0.0002 0.0000 0.0002	-0.0004 0.0000 -0.0002	-0.0001 0.0000 -0.0001	-0.0001 0.0000 -0.0001	-0.0001 0.0000 -0.0001	-0.0539 -0.9027 -0.9027	0.6281 1.1733 1.1733	12.05 0.0508 0.0537	0.0013 0.0011 0.0011	0.0067 0.0070 0.0070	0.0067 0.0070 0.0070	0.1336 0.1063 0.1063	0.6643 0.6556 0.6556

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																									
			1	2	3	4	5	6	7	8	9	10	11	12	13	14	15	16	17	18	19	20						
489	2	7	0.898	-0.98	-0.00	-0.01	12.07	0.06	12.08	0.1498	0.6210	0.0048	0.0000	-0.0006	0.0003	0.6481	0.0674	0.0020	0.0083	0.1145	0.6205	-0.0008	0.0002	-0.0002	1.6179	0.0704	0.0016	0.0087
489	3	7	0.896	0.04	-0.00	-0.01	12.09	0.06	12.08	0.1587	0.6085	0.0056	-0.0000	0.0007	0.6343	0.0738	0.0023	0.0089	0.1587	0.6103	-0.0012	0.0003	-0.0002	1.0427	0.0763	0.0018	0.0093	
489	4	7	0.897	0.55	-0.00	-0.01	12.09	0.06	12.09	0.1543	0.6048	0.0057	-0.0000	0.0007	0.6263	0.0780	0.0024	0.0094	0.1543	0.6062	-0.0012	0.0003	-0.0002	1.0639	0.0794	0.0019	0.0096	
489	5	7	0.895	1.08	-0.00	-0.01	12.09	0.06	12.10	0.1520	0.5988	0.0059	-0.0000	0.0007	0.6107	0.0818	0.0024	0.0098	0.1520	0.5997	-0.0009	0.0003	-0.0003	1.5314	0.0824	0.0021	0.0098	
489	6	7	0.897	3.16	-0.00	-0.01	12.11	0.06	12.10	0.1342	0.5885	0.0050	-0.0000	0.0006	0.6307	0.0962	0.0025	0.0113	0.1342	0.5834	-0.0013	0.0003	-0.0002	1.1078	0.0955	0.0021	0.0111	
489	7	7	0.896	6.26	-0.00	-0.02	12.12	0.06	12.12	0.1118	0.5775	0.0038	-0.0001	0.0004	0.5925	0.1075	0.0024	0.0124	0.1118	0.5728	-0.0006	0.0003	-0.0001	1.4056	0.1107	0.0019	0.0126	
489	8	7	0.896	9.31	-0.00	-0.02	12.12	0.06	12.11	0.0865	0.5750	0.0031	-0.0000	0.0004	0.6506	0.1148	0.0015	0.0132	0.0865	0.5707	-0.0006	0.0002	-0.0002	1.4777	0.1179	0.0015	0.0134	
489	9	7	0.898	12.49	0.00	-0.02	12.12	0.06	12.09	0.0822	0.5698	0.0032	-0.0001	0.0004	0.6250	0.1208	0.0015	0.0138	0.0822	0.5660	-0.0014	0.0002	-0.0002	0.9571	0.1227	0.0015	0.0138	
489	10	7	0.898	0.03	-0.00	-0.02	12.09	0.06	12.09	0.1555	0.6054	0.0058	-0.0000	0.0006	0.5807	0.0740	0.0018	0.0089	0.1555	0.6095	-0.0015	0.0003	-0.0002	0.7778	0.0766	0.0018	0.0093	
490	1	7	0.897	3.00	-0.00	-0.02	15.09	0.06	15.07	0.1436	0.6226	0.0039	-0.0000	0.0004	0.6141	0.0668	0.0015	0.0083	0.1436	0.6226	-0.0013	0.0002	-0.0003	1.2260	0.0698	0.0015	0.0087	
490	2	7	0.900	-0.98	-0.00	-0.02	15.10	0.06	15.09	0.1482	0.6040	0.0050	-0.0000	0.0006	0.6571	0.0816	0.0024	0.0098	0.1482	0.6009	-0.0009	0.0002	-0.0002	1.3530	0.0835	0.0020	0.0100	

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Pt. Cd	See Key																								
490	5	0.899	0.000	0.03	-0.00	-0.00	-0.00	-0.00	-0.01	15.11	0.06	15.09	0.1450	0.5923	0.0057	0.000	0.000	0.000	0.000	0.6122	0.0877	0.0025	0.0103	0.1252	0.5898	
	6	-0.0011	0.0003	-0.0003	0.0002	-0.0002	-0.0002	-0.0002	1.2064	0.0885	0.0022	0.0104														
490	4	0.898	0.000	0.58	-0.00	-0.00	-0.00	-0.00	-0.02	15.12	0.06	15.05	0.1396	0.5909	0.0060	0.000	0.000	0.000	0.000	0.6041	0.0914	0.0025	0.0108	0.1183	0.5865	
	6	-0.0013	0.0003	-0.0003	0.0002	-0.0002	-0.0002	-0.0002	0.8957	0.0918	0.0021	0.0107														
490	5	0.899	1.09	0.000	-0.00	-0.00	-0.00	-0.00	-0.02	15.12	0.06	15.10	0.1344	0.5879	0.0060	0.000	0.000	0.000	0.000	0.6050	0.0954	0.0025	0.0112	0.1106	0.5840	
	6	-0.0014	0.0003	-0.0003	0.0002	-0.0002	-0.0002	-0.0002	0.9395	0.0950	0.0021	0.0111														
490	6	0.899	3.16	0.000	-0.00	-0.00	-0.00	-0.00	-0.02	15.13	0.06	15.10	0.1219	0.5759	0.0052	0.000	0.000	0.000	0.000	0.6287	0.1036	0.0025	0.0119	0.1027	0.5769	
	6	-0.0013	0.0003	-0.0003	0.0002	-0.0002	-0.0002	-0.0002	1.0223	0.1055	0.0021	0.0121														
490	7	0.900	6.26	0.000	-0.00	-0.00	-0.00	-0.00	-0.02	15.13	0.06	15.09	0.0921	0.5759	0.0038	0.000	0.000	0.000	0.000	0.6360	0.1131	0.0020	0.0130	0.0716	0.5714	
	6	-0.0006	0.0003	-0.0003	0.0001	-0.0001	-0.0001	-0.0001	1.2261	0.1148	0.0016	0.0131														
490	8	0.898	9.33	0.000	-0.00	-0.00	-0.00	-0.00	-0.02	15.13	0.06	15.10	0.0857	0.5690	0.0038	0.000	0.000	0.000	0.000	0.6200	0.1193	0.0020	0.0121	0.0644	0.5697	
	6	-0.0006	0.0002	-0.0002	0.0001	-0.0001	-0.0001	-0.0001	1.2779	0.1214	0.0015	0.0121														
490	9	0.898	12.48	0.000	-0.00	-0.00	-0.00	-0.00	-0.02	15.13	0.06	15.10	0.0733	0.5703	0.0037	0.000	0.000	0.000	0.000	0.5812	0.1254	0.0018	0.0143	0.0511	0.5697	
	6	-0.0014	0.0002	-0.0002	0.0002	-0.0002	-0.0002	-0.0002	0.9722	0.1269	0.0012	0.0144														
490	10	0.898	0.05	0.000	-0.00	-0.00	-0.00	-0.00	-0.01	15.11	0.06	15.09	0.1445	0.5893	0.0059	0.000	0.000	0.000	0.000	0.5943	0.0879	0.0025	0.0103	0.1234	0.5875	
	6	-0.0015	0.0003	-0.0003	0.0002	-0.0002	-0.0002	-0.0002	0.8241	0.0891	0.0022	0.0104														
491	3	1.248	-3.24	0.000	11.20	-0.0025	-0.0007	-0.0007	-3.02	-0.0261	-0.0004	-0.0036	0.0939	0.6941	0.0190	0.0005	0.0001	0.0001	0.6580	-0.0228	0.05	-0.0031	0.1354	0.6910		
	6	-0.0052	0.0001	-0.0001	0.0007	-0.0007	-0.0007	-0.0007	0.7147	-0.0228	-0.0006	-0.0031														
491	4	1.249	-3.19	0.000	22.50	-0.0032	-0.0012	-0.0012	-3.03	-0.0245	0.04	-0.0034	0.0953	0.6971	0.0247	0.0006	0.0006	0.0006	0.6629	-0.0212	0.04	-0.0029	0.1429	0.6967		
	6	-0.0100	0.0000	-0.0000	0.0012	-0.0012	-0.0012	-0.0012	0.6392	-0.0212	-0.0006	-0.0029														
491	5	1.249	-3.18	0.000	33.70	-0.0039	-0.0018	-0.0018	-3.04	-0.0217	0.03	-0.0030	0.1011	0.7007	0.0299	0.0006	0.0006	0.0006	0.6669	-0.0217	0.03	-0.0030	0.1501	0.7079		
	6	-0.0147	0.0000	-0.0000	0.0018	-0.0018	-0.0018	-0.0018	0.6240	-0.0185	0.03	-0.0026														

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																		
491	6	1.248	-3.19	45.00	0.0964	-3.04	-0.0177	0.02	-0.0024	0.1177	0.7041	-0.0348	-0.0006	-0.0046	0.6656	-0.0146	-0.0004	-0.0021	0.1658	0.7220
	7	-0.0192	-0.0000	-0.0025	0.0246	0.6541	-0.0146	-0.0004	-0.0021	0.1658	0.7220	-0.0192	-0.0000	-0.0025	0.6541	-0.0146	-0.0004	-0.0021	0.1658	0.7220
491	7	1.248	-3.19	56.20	0.0924	-3.05	-0.0134	0.02	-0.0019	0.1461	0.7140	-0.0383	-0.0007	-0.0051	0.6666	-0.0105	-0.0004	-0.0015	0.1917	0.7399
	7	-0.0230	-0.0001	-0.0030	0.0291	0.6520	-0.0105	-0.0004	-0.0015	0.1917	0.7399	-0.0230	-0.0001	-0.0030	0.6520	-0.0105	-0.0004	-0.0015	0.1917	0.7399
491	8	1.249	-3.19	67.50	0.0693	-3.05	-0.0092	0.01	-0.0013	0.1783	0.7205	-0.0412	-0.0007	-0.0055	0.6678	-0.0061	-0.0003	-0.0009	0.2674	0.7620
	7	-0.0262	-0.0001	-0.0034	0.0331	0.6551	-0.0061	-0.0003	-0.0009	0.2674	0.7620	-0.0262	-0.0001	-0.0034	0.6551	-0.0061	-0.0003	-0.0009	0.2674	0.7620
491	9	1.248	-3.20	78.69	0.0873	-3.05	-0.0046	0.01	-0.0007	0.2935	0.8519	-0.0428	-0.0007	-0.0057	0.6648	-0.0018	-0.0002	-0.0003	0.6606	1.0532
	7	-0.0280	-0.0001	-0.0036	0.0350	0.6519	-0.0018	-0.0002	-0.0003	0.6606	1.0532	-0.0280	-0.0001	-0.0036	0.6519	-0.0018	-0.0002	-0.0003	0.6606	1.0532
491	10	1.249	-3.19	89.99	0.0852	-3.05	-0.002	0.01	-0.0001	-3.4439	-2.8932	-0.0435	-0.0007	-0.0058	0.6670	-0.0018	-0.0001	-0.0001	-0.3504	0.3386
	7	-0.0287	-0.0002	-0.0037	0.0377	0.6535	-0.002	-0.0001	-0.0001	-3.4439	-2.8932	-0.0287	-0.0002	-0.0037	0.6535	-0.0018	-0.0001	-0.0001	-0.3504	0.3386
492	1	1.249	-0.05	11.20	0.15331	-3.01	-0.006	0.05	-0.0002	-0.1146	0.5131	-0.0124	-0.0004	-0.0016	0.6447	-0.0021	-0.0001	-0.0002	-0.3875	0.5457
	7	-0.0009	-0.0002	-0.0001	-1.5331	0.6665	-0.006	-0.0001	-0.0002	-0.1146	0.5457	-0.0009	-0.0002	-0.0001	0.6665	-0.0021	-0.0001	-0.0002	-0.3875	0.5457
492	2	1.249	-0.04	22.49	0.1629	-3.01	-0.006	0.05	-0.0002	-0.2045	0.3678	-0.0127	-0.0004	-0.0016	0.6423	-0.0021	-0.0001	-0.0002	-0.3752	0.5157
	7	-0.0011	-0.0003	-0.0001	-1.2998	0.5972	-0.006	-0.0001	-0.0002	-0.2045	0.5157	-0.0011	-0.0003	-0.0001	0.5972	-0.0021	-0.0001	-0.0002	-0.3752	0.5157
492	3	1.249	-0.05	33.69	0.1775	-3.01	-0.006	0.06	-0.0002	-0.3820	0.1246	-0.0130	-0.0004	-0.0016	0.6366	-0.0022	-0.0001	-0.0002	-0.3518	0.4584
	7	-0.0014	-0.0003	-0.0001	-1.0503	0.6798	-0.006	-0.0002	-0.0002	-0.3820	0.4584	-0.0014	-0.0003	-0.0001	0.6798	-0.0022	-0.0001	-0.0002	-0.3518	0.4584
492	4	1.249	-0.05	44.99	0.1689	-3.01	-0.006	0.05	-0.0002	-0.8940	-0.3160	-0.0138	-0.0004	-0.0017	0.6338	-0.0022	-0.0001	-0.0002	-0.3814	0.4770
	7	-0.0014	-0.0003	-0.0001	-1.0191	0.6534	-0.006	-0.0002	-0.0002	-0.8940	-0.3160	-0.0014	-0.0003	-0.0001	0.6534	-0.0022	-0.0001	-0.0002	-0.3814	0.4770
492	5	1.250	-0.05	56.19	0.1657	-3.02	-0.007	0.06	-0.0002	-28.6305	-22.7320	-0.0145	-0.0004	-0.0018	0.6346	-0.0023	-0.0001	-0.0001	-0.3977	0.3994
	7	-0.0016	-0.0003	-0.0001	-0.9804	0.5273	-0.007	-0.0002	-0.0001	-28.6305	-22.7320	-0.0016	-0.0003	-0.0001	0.5273	-0.0023	-0.0001	-0.0001	-0.3977	0.3994
492	6	1.249	-0.06	67.49	0.1603	-3.02	-0.007	0.06	-0.0002	-1.4525	1.7789	-0.0150	-0.0004	-0.0019	0.6342	-0.0022	-0.0001	-0.0001	-0.4206	0.3809
	7	-0.0015	-0.0003	-0.0001	-1.1243	0.5968	-0.007	-0.0002	-0.0001	-1.4525	1.7789	-0.0015	-0.0003	-0.0001	0.5968	-0.0022	-0.0001	-0.0001	-0.4206	0.3809

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key													
492	7	7	1.249	-0.0151	-0.0004	-0.0003	78.69	0.1608	-3.02	-0.07	0.06	-0.003	0.7957	1.3762		
	6	5	-0.0014	0.0004	0.0003	-0.0019	-1.2561	0.5550	0.0021	-0.0001	0.0001	-0.4288	0.3516			
492	7	9	1.249	-0.0152	-0.0004	-0.0003	89.99	0.1571	-3.02	-0.06	0.06	1.1362	1.5337			
	6	9	-0.0013	0.0003	0.0003	-0.0000	-1.3712	0.3305	0.0020	-0.0001	0.0001	-0.4760	0.2936			
493	7	5	1.249	-0.0099	-0.0003	-0.0003	11.19	0.1870	-3.01	-0.05	0.06	0.0855	0.6708			
	6	5	0.0001	0.0003	0.0003	0.0000	9.7068	1.4898	0.0088	0.0000	0.0011	0.0246	0.6638			
493	7	9	1.250	-0.0089	-0.0003	-0.0004	22.49	0.1854	-3.00	-0.05	0.06	0.0944	0.6762			
	6	9	0.0013	0.0004	0.0004	0.0002	1.6829	0.7853	0.0079	0.0000	0.0011	0.0211	0.6522			
493	7	9	1.251	-0.0079	-0.0003	-0.0004	33.69	0.2205	-3.00	-0.05	0.07	0.0812	0.6608			
	6	9	0.0023	0.0005	0.0005	0.0004	1.0679	0.8444	0.0078	0.0000	0.0010	0.0228	0.6563			
493	7	9	1.250	-0.0039	-0.0003	-0.0005	44.99	0.2412	-3.00	-0.05	0.07	0.0152	0.6199			
	6	9	0.0075	0.0003	0.0003	0.0005	0.6688	0.7163	0.0049	0.0000	0.0008	0.0056	0.6224			
493	7	9	1.250	-0.0072	-0.0003	-0.0005	56.19	0.2577	-3.00	-0.06	0.07	-0.0905	0.5595			
	6	9	0.0051	0.0005	0.0005	0.0007	0.5620	0.6988	0.0056	-0.0000	0.0007	-0.0079	0.6272			
493	7	9	1.249	-0.0073	-0.0003	-0.0006	67.49	0.2527	-3.00	-0.06	0.07	-0.3056	0.2490			
	6	9	0.0059	0.0006	0.0006	0.0008	0.5079	0.7263	0.0046	-0.0000	0.0005	-0.0977	0.6111			
493	7	9	1.249	-0.0071	-0.0003	-0.0006	78.69	0.2517	-3.00	-0.07	0.07	-2.5176	-1.7159			
	6	9	0.0067	0.0006	0.0006	0.0009	0.4512	0.7183	0.0033	-0.0001	0.0003	-0.2242	0.5033			
493	7	9	1.250	-0.0071	-0.0003	-0.0005	89.99	0.2390	-3.00	-0.07	0.07	-1.1018	1.6831			
	6	9	0.0071	0.0003	0.0003	0.0008	0.4043	0.6777	0.0020	-0.0002	0.0000	-0.5038	0.2062			
494	7	9	1.248	-0.0093	-0.0008	-0.0008	89.99	-0.0236	-2.97	-0.07	0.11	0.9423	1.3978			
	6	9	0.0253	0.0000	0.0000	0.0034	0.1714	0.6811	0.0016	-0.0002	-0.0000	-0.6982	0.2295			

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key													
494	2	1.249 0.0093 0.0259	5.11 0.0000 0.0000	78.69 0.0012 0.0035	-0.0176 0.1695	-2.97 0.6871 0.6834	-0.06 0.0032 0.0058	0.11 0.0000 0.0000	-0.001 0.0003 0.0006	-0.1463 0.0745	0.5503 0.5922				
494	3	1.249 0.0085 0.0248	3.11 0.0000 0.0008	67.49 0.0012 0.0034	0.0007 0.1805	-2.97 0.7194 0.6880	-0.05 0.0074 0.0102	0.10 0.0000 0.0000	-0.000 0.0013 0.0013	0.0306 0.0313	0.6357 0.6531				
494	4	1.248 0.0069 0.0223	5.11 0.0000 0.0008	56.19 0.0009 0.0030	0.0010 0.1982	-2.97 0.6977 0.6818	-0.04 0.0120 0.0150	0.10 0.0001 0.0001	-0.000 0.0015 0.0018	0.0511 0.0486	0.6471 0.6302				
494	5	1.248 0.0042 0.0182	3.11 0.0000 0.0008	44.99 0.0006 0.0024	-0.0370 0.2351	-2.98 0.7215 0.6835	-0.03 0.0167 0.0190	0.10 0.0002 0.0002	0.000 0.0021 0.0024	0.0657 0.0603	0.6520 0.6404				
494	6	1.250 0.0008 0.0136	3.12 0.0001 0.0007	33.69 0.0001 0.0017	-0.7009 0.2859	-2.99 0.8001 0.6548	-0.03 0.0210 0.0224	0.09 0.0002 0.0002	0.000 0.0029 0.0029	0.0677 0.0632	0.6593 0.6547				
494	7	1.249 0.0025 0.0083	3.12 0.0002 0.0006	22.49 0.0002 0.0011	0.4629 0.4138	-2.99 0.5523 0.6842	-0.02 0.0248 0.0247	0.08 0.0003 0.0002	0.001 0.0032 0.0032	0.0674 0.0564	0.6496 0.6584				
494	8	1.251 0.0065 0.0032	5.13 0.0003 0.0005	11.19 0.0008 0.0004	0.2540 0.8646	-3.00 0.6429 0.7586	-0.01 0.0273 0.0263	0.07 0.0003 0.0002	0.000 0.0036 0.0034	0.0714 0.0473	0.6574 0.6532				
495	1	1.247 0.0026 0.0092	6.30 0.0002 0.0006	11.19 0.0003 0.0012	0.4164 0.3682	-2.99 0.6139 0.6778	0.01 0.0548 0.0569	0.08 0.0006 0.0004	0.003 0.0073 0.0075	0.0611 0.0354	0.6661 0.6645				
495	2	1.249 0.0060 0.0210	6.27 0.0000 0.0008	22.49 0.0007 0.0028	-0.0043 0.2029	-2.98 0.6293 0.6759	0.01 0.0507 0.0533	0.10 0.0006 0.0003	0.003 0.0067 0.0071	0.0601 0.0371	0.6637 0.6674				
495	3	1.250 0.0149 0.0318	6.27 0.0001 0.0009	33.70 0.0019 0.0044	0.0502 0.1475	-2.96 0.6590 0.6919	0.00 0.0449 0.0476	0.12 0.0005 0.0005	0.003 0.0059 0.0063	0.0610 0.0414	0.6627 0.6678				
495	4	1.249 0.0232 0.0412	6.27 0.0002 0.0009	44.99 0.0031 0.0056	0.0525 0.1209	-2.95 0.6668 0.6898	-0.01 0.0372 0.0402	0.13 0.0004 0.0003	0.001 0.0049 0.0053	0.0636 0.0431	0.6577 0.6596				

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key													
495	5	1.251	6.28	56.19	0.0558	-2.94	-0.02	0.14	0.01	0.0658	0.6473	0.0301	0.0036	0.0521	0.6546
	8	0.0301	0.0003	0.0040	0.1095	0.6717	0.0308	0.0003	0.0036	0.0521	0.6546	0.0003	0.0040	0.0521	0.6546
	9	0.0478	0.0010	0.0065	0.1095	0.6831	0.0308	0.0003	0.0040	0.0521	0.6546	0.0003	0.0040	0.0521	0.6546
495	6	1.250	6.28	67.49	0.0553	-2.93	-0.03	0.15	0.00	0.0628	0.6311	0.0002	0.0023	0.0598	0.6395
	8	0.0351	0.0003	0.0047	0.1031	0.6771	0.0185	0.0002	0.0023	0.0598	0.6395	0.0002	0.0025	0.0598	0.6395
	9	0.0517	0.0010	0.0070	0.1031	0.6784	0.0201	0.0002	0.0025	0.0598	0.6395	0.0002	0.0025	0.0598	0.6395
495	7	1.250	6.28	78.69	0.0487	-2.93	-0.05	0.15	-0.00	0.0387	0.5849	0.0000	0.0010	0.0228	0.6049
	8	0.0386	0.0003	0.0052	0.0980	0.6776	0.0085	0.0000	0.0010	0.0228	0.6049	0.0000	0.0011	0.0228	0.6049
	9	0.0539	0.0010	0.0072	0.0980	0.6750	0.0095	0.0000	0.0011	0.0228	0.6049	0.0000	0.0011	0.0228	0.6049
495	8	1.250	6.29	89.99	0.0416	-2.93	-0.07	0.15	-0.02	3.3437	3.7454	-0.0002	-0.0002	-3.5357	-1.3534
	8	0.0395	0.0003	0.0053	0.0982	0.6788	-0.0002	0.0001	-0.0002	3.3437	3.7454	-0.0002	-0.0002	-3.5357	-1.3534
	9	0.0539	0.0010	0.0072	0.0982	0.6742	0.0003	0.0001	-0.0002	3.3437	3.7454	-0.0002	-0.0002	-3.5357	-1.3534
496	3	1.248	9.44	11.20	-0.2845	-2.99	0.05	0.08	0.06	0.0656	0.6534	0.0010	0.0102	0.0378	0.6478
	8	0.0017	-0.0001	0.0002	0.1978	0.7878	0.0785	0.0010	0.0102	0.0656	0.6534	0.0010	0.0102	0.0378	0.6478
	9	0.0148	0.0005	0.0019	0.1978	0.6660	0.0845	0.0006	0.0102	0.0656	0.6534	0.0006	0.0109	0.0378	0.6478
496	4	1.249	9.42	22.50	0.0361	-2.96	0.04	0.11	0.06	0.0645	0.6569	0.0009	0.0096	0.0387	0.6497
	8	0.0159	0.0001	0.0021	0.1063	0.6618	0.0733	0.0009	0.0096	0.0645	0.6569	0.0009	0.0096	0.0387	0.6497
	9	0.0336	0.0007	0.0046	0.1063	0.6884	0.0810	0.0006	0.0096	0.0645	0.6569	0.0006	0.0105	0.0387	0.6497
496	5	1.248	9.43	33.70	0.0532	-2.94	0.03	0.14	0.04	0.0663	0.6617	0.0008	0.0086	0.0405	0.6533
	8	0.0296	0.0003	0.0039	0.0934	0.6739	0.0655	0.0008	0.0086	0.0663	0.6617	0.0008	0.0086	0.0405	0.6533
	9	0.0499	0.0009	0.0067	0.0934	0.6775	0.0745	0.0006	0.0097	0.0663	0.6617	0.0006	0.0097	0.0405	0.6533
496	6	1.249	9.42	44.99	0.0544	-2.93	0.02	0.15	0.04	0.0691	0.6696	0.0007	0.0073	0.0401	0.6613
	8	0.0426	0.0004	0.0057	0.0877	0.6726	0.0546	0.0007	0.0073	0.0691	0.6696	0.0007	0.0073	0.0401	0.6613
	9	0.0626	0.0010	0.0083	0.0877	0.6653	0.0645	0.0005	0.0085	0.0691	0.6696	0.0005	0.0085	0.0401	0.6613
496	7	1.249	9.42	56.18	0.0526	-2.91	-0.00	0.17	0.01	0.0709	0.6709	0.0003	0.0056	0.0355	0.6615
	8	0.0529	0.0005	0.0071	0.0852	0.6758	0.0421	0.0003	0.0056	0.0709	0.6709	0.0003	0.0056	0.0355	0.6615
	9	0.0713	0.0012	0.0093	0.0852	0.6568	0.0515	0.0003	0.0068	0.0709	0.6709	0.0003	0.0068	0.0355	0.6615
496	8	1.250	9.43	67.48	0.0483	-2.90	-0.02	0.18	0.00	0.0727	0.6590	0.0004	0.0036	0.0347	0.6602
	8	0.0607	0.0005	0.0080	0.0830	0.6669	0.0276	0.0004	0.0036	0.0727	0.6590	0.0004	0.0036	0.0347	0.6602
	9	0.0770	0.0012	0.0100	0.0830	0.6512	0.0343	0.0002	0.0045	0.0727	0.6590	0.0002	0.0045	0.0347	0.6602
496	9	1.249	9.43	78.69	0.0458	-2.89	-0.04	0.18	-0.01	0.0776	0.6388	0.0001	0.0016	0.0434	0.6459
	8	0.0658	0.0006	0.0087	0.0804	0.6641	0.0126	0.0001	0.0016	0.0776	0.6388	0.0001	0.0016	0.0434	0.6459
	9	0.0799	0.0012	0.0103	0.0804	0.6468	0.0160	0.0001	0.0020	0.0776	0.6388	0.0001	0.0020	0.0434	0.6459

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																				
496	10	7	1.250	2.43	89.99	0.0444	-2.88	-0.07	0.19	-0.04	0.9443	1.5803	0.0685	0.0006	0.0090	0.0783	0.6619	-0.0007	-0.0001	-0.0002	-1.9459	-1.0060	
		6	0.0799	0.0012	0.0103	0.0783	0.6455	0.0005	-0.0001	-0.0001	-0.0001	-1.0060											
497	1	7	1.249	12.64	89.99	0.0481	-2.85	-0.06	0.22	-0.03	-4.6166	-2.0655	0.0923	0.0008	0.0118	0.0739	0.6419	0.0001	-0.0001	-0.0001	-0.8744	-0.0291	
		6	0.1004	0.0014	0.0125	0.0739	0.6245	0.0011	-0.0002	-0.0002	-0.0002	-0.0291											
497	2	7	1.250	12.63	78.68	0.0460	-2.85	-0.03	0.22	-0.00	0.0747	0.6512	0.0890	0.0008	0.0114	0.0751	0.6417	0.0002	0.0001	0.0001	0.0286	0.6509	
		6	0.1008	0.0015	0.0125	0.0751	0.6230	0.0234	0.0001	0.0001	0.0001	0.6512											
497	3	7	1.250	12.63	67.47	0.0486	-2.86	-0.00	0.22	0.02	0.0750	0.6740	0.0822	0.0008	0.0106	0.0771	0.6490	0.0005	0.0003	0.0005	0.0362	0.6638	
		6	0.0981	0.0015	0.0122	0.0771	0.6249	0.0491	0.0003	0.0003	0.0003	0.6638											
497	4	7	1.250	12.61	56.18	0.0521	-2.88	0.02	0.21	0.03	0.0699	0.6572	0.0730	0.0007	0.0095	0.0795	0.6563	0.0008	0.0006	0.0009	0.0434	0.6574	
		6	0.0920	0.0014	0.0116	0.0795	0.6345	0.0692	0.0006	0.0006	0.0006	0.6574											
497	5	7	1.250	12.61	44.99	0.0562	-2.89	0.04	0.18	0.06	0.0653	0.6464	0.0605	0.0006	0.0080	0.0854	0.6632	0.0009	0.0007	0.0009	0.0436	0.6507	
		6	0.0812	0.0013	0.0104	0.0854	0.6421	0.0848	0.0007	0.0007	0.0007	0.6464											
497	6	7	1.249	12.62	33.70	0.0653	-2.92	0.05	0.16	0.07	0.0617	0.6365	0.0442	0.0005	0.0059	0.0932	0.6698	0.0008	0.0008	0.0122	0.0440	0.6489	
		6	0.0660	0.0012	0.0086	0.0932	0.6556	0.0960	0.0008	0.0008	0.0122	0.6365											
497	7	7	1.250	12.64	22.51	0.0706	-2.93	0.07	0.14	0.08	0.0605	0.6398	0.0256	0.0003	0.0034	0.1030	0.6694	0.0008	0.0008	0.0130	0.0418	0.6316	
		6	0.0465	0.0009	0.0062	0.1030	0.6694	0.1033	0.0008	0.0008	0.0130	0.6398											
497	8	7	1.250	12.64	11.20	0.0297	-2.98	0.08	0.09	0.08	0.0607	0.6231	0.0060	0.0006	0.0008	0.1522	0.6481	0.0008	0.0008	0.0133	0.0402	0.6231	
		6	0.0208	0.0006	0.0027	0.1522	0.6481	0.1066	0.0008	0.0008	0.0133	0.6231											
497	9	7	1.249	12.65	11.20	0.0446	0.00	0.07	0.09	0.08	0.0606	0.6331	0.0212	0.0001	0.0028	0.0446	0.6691	0.0008	0.0008	0.0133	0.0402	0.6231	
		6	0.0212	0.0001	0.0028	0.0446	0.6691	0.1066	0.0008	0.0008	0.0133	0.6331											
498	1	7	1.251	12.66	11.20	0.0447	0.00	0.08	0.10	0.07	0.0603	0.6320	0.0211	0.0001	0.0028	0.0447	0.6707	0.0008	0.0008	0.0133	0.0401	0.6244	
		6	0.0213	0.0006	0.0028	0.0447	0.6606	0.1064	0.0008	0.0008	0.0133	0.6320											

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																				
498	2	1.249	12.65	22.51	0.0511	0.6787	0.06	0.14	0.08	0.0600	0.6392	0.0419	0.0004	0.0056	0.0063	0.1077	0.6751	0.1033	0.0008	0.0118	0.0417	0.6319
	6	0.0470	0.0010	0.0056	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
	9	0.0470	0.0010	0.0056	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
498	3	1.250	12.62	33.71	0.0495	0.6694	0.05	0.17	0.06	0.0605	0.6449	0.0598	0.0005	0.0080	0.0087	0.0958	0.6611	0.0958	0.0108	0.0445	0.6382	0.0661
	6	0.0598	0.0005	0.0080	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
	9	0.0661	0.0012	0.0087	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
498	4	1.249	12.63	44.99	0.0467	0.6571	0.03	0.19	0.06	0.0632	0.6521	0.0753	0.0007	0.0099	0.0105	0.0674	0.6462	0.0847	0.0094	0.0428	0.6461	0.0814
	6	0.0753	0.0007	0.0099	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
	9	0.0814	0.0014	0.0105	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
498	5	1.249	12.64	56.18	0.0457	0.6461	0.02	0.21	0.04	0.0685	0.6635	0.0869	0.0005	0.0112	0.0116	0.0804	0.6344	0.0689	0.0075	0.0425	0.6573	0.0921
	6	0.0869	0.0007	0.0112	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
	9	0.0921	0.0014	0.0116	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
498	6	1.249	12.65	67.48	0.0443	0.6384	-0.01	0.22	0.02	0.0722	0.6627	0.0983	0.0003	0.0122	0.0122	0.0777	0.6242	0.0484	0.0051	0.0372	0.6657	0.0957
	6	0.0957	0.0008	0.0122	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
	9	0.0983	0.0015	0.0122	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
498	7	1.251	12.65	78.69	0.0437	0.6308	-0.04	0.22	0.00	0.0711	0.6343	0.1019	0.0001	0.0128	0.0125	0.0759	0.6220	0.0487	0.0023	0.0311	0.6550	0.1010
	6	0.1019	0.0008	0.0128	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
	9	0.1010	0.0015	0.0125	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
498	8	1.248	12.65	89.99	0.0470	0.6308	-0.07	0.22	0.03	1.0218	2.3780	0.1045	0.0002	0.0131	0.0125	0.0745	0.6229	0.0004	0.0002	1.2355	0.1098	0.1005
	6	0.1045	0.0009	0.0131	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
	9	0.1005	0.0015	0.0125	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
499	1	1.249	9.45	89.99	0.0452	0.6535	-0.07	0.19	0.03	1.0251	1.7531	0.0836	0.0001	0.0109	0.0103	0.0796	0.6432	0.0005	0.0001	-1.8041	0.9729	0.0801
	6	0.0836	0.0007	0.0109	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
	9	0.0801	0.0012	0.0103	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
499	2	1.249	9.45	78.69	0.0437	0.6533	-0.05	0.19	0.01	0.0612	0.6388	0.0812	0.0001	0.0106	0.0103	0.0803	0.6422	0.0128	0.0016	0.0410	0.6241	0.0804
	6	0.0812	0.0007	0.0106	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
	9	0.0804	0.0012	0.0103	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
499	3	1.250	9.44	67.49	0.0430	0.6572	-0.03	0.18	0.00	0.0690	0.6593	0.0762	0.0002	0.0100	0.0100	0.0829	0.6462	0.0276	0.0036	0.0339	0.6574	0.0774
	6	0.0762	0.0006	0.0100	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
	9	0.0774	0.0012	0.0100	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
499	4	1.249	9.44	56.19	0.0422	0.6622	-0.00	0.17	0.02	0.0646	0.6668	0.0688	0.0003	0.0091	0.0093	0.0848	0.6491	0.0424	0.0056	0.0346	0.6612	0.0718
	6	0.0688	0.0005	0.0091	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
	9	0.0718	0.0012	0.0093	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key													
499	5	7	1.250	9.44	44.99	0.0437	0.06	0.01	0.16	0.03	0.0636	0.0648	0.0636	0.0401	0.6623
		9	0.0589	0.0005	0.0078	0.0871	0.6695	0.552	0.0007	0.0073					
		7	0.0631	0.0010	0.0082				0.0005	0.0085					
499	6	7	1.249	9.44	33.70	0.0409	0.04	0.02	0.14	0.04	0.0623	0.0687	0.0623	0.0394	0.6505
		9	0.0463	0.0003	0.0062	0.0920	0.6760	0.659	0.0008	0.0086					
		5	0.0505	0.0009	0.0067				0.0005	0.0097					
499	7	7	1.248	9.44	22.50	0.0377	0.02	0.03	0.11	0.05	0.0604	0.0619	0.0604	0.0382	0.6519
		9	0.0322	0.0002	0.0043	0.1043	0.6704	0.737	0.0008	0.0096					
		9	0.0342	0.0007	0.0045				0.0006	0.0105					
499	8	7	1.248	9.45	11.20	0.0239	-0.00	0.04	0.09	0.05	0.0621	0.0637	0.0621	0.0368	0.6507
		9	0.0164	0.0000	0.0021	0.2035	0.6613	0.788	0.0009	0.0102					
		9	0.0151	0.0006	0.0019				0.0006	0.0109					
500	1	7	1.248	6.28	11.20	0.0119	-0.00	0.01	0.07	0.03	0.0644	0.0654	0.0644	0.0350	0.6654
		9	0.0117	0.0000	0.0015	0.3031	0.6516	0.544	0.0007	0.0072					
		9	0.0100	0.0006	0.0012				0.0004	0.0076					
500	2	7	1.250	6.29	22.50	0.0504	0.01	0.01	0.09	0.02	0.0642	0.0659	0.0642	0.0371	0.6673
		9	0.0219	0.0002	0.0029	0.1741	0.6679	0.502	0.0006	0.0066					
		9	0.0218	0.0007	0.0028				0.0004	0.0071					
500	3	7	1.250	6.29	33.70	0.0502	0.02	-0.00	0.11	0.02	0.0647	0.0650	0.0647	0.0400	0.6666
		9	0.0319	0.0003	0.0043	0.1243	0.6753	0.444	0.0005	0.0059					
		9	0.0328	0.0008	0.0044				0.0003	0.0064					
500	4	7	1.250	6.28	44.99	0.0443	0.03	-0.01	0.13	0.01	0.0693	0.0629	0.0693	0.0452	0.6641
		9	0.0409	0.0003	0.0055	0.1044	0.6770	0.366	0.0005	0.0048					
		9	0.0420	0.0008	0.0056				0.0003	0.0053					
500	5	7	1.250	6.28	56.19	0.0483	0.04	-0.02	0.14	0.00	0.0762	0.0658	0.0762	0.0536	0.6555
		9	0.0475	0.0004	0.0064	0.0942	0.6812	0.275	0.0004	0.0036					
		9	0.0487	0.0009	0.0065				0.0003	0.0040					
500	6	7	1.250	6.29	67.49	0.0459	0.05	-0.04	0.14	-0.00	0.0737	0.0638	0.0737	0.0540	0.6382
		9	0.0529	0.0004	0.0071	0.0876	0.6745	0.181	0.0002	0.0023					
		9	0.0529	0.0009	0.0070				0.0002	0.0026					
500	7	7	1.249	6.29	78.69	0.0434	0.05	-0.06	0.15	-0.01	0.0825	0.0623	0.0825	0.0234	0.6233
		9	0.0562	0.0004	0.0076	0.0868	0.6774	0.080	0.0001	0.0010					
		9	0.0546	0.0009	0.0072				0.0000	0.0012					

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd																					
500	8	1.250	6.29	89.99	0.0404	0.06	-0.08	0.15	-0.003	0.08	-0.0002	0.001	-0.0002	0.0002	0.6776	1.5705						
	8	0.8576	0.0004	0.0077	0.00877	0.6748	-0.0008	-0.0001	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-3.4924	-1.1186						
	9	0.0543	0.0009	0.0072	0.00877	0.6676	0.0003	-0.0002	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	0.7467	1.3046						
501	1	1.245	3.14	89.99	0.0401	0.01	-0.07	0.10	-0.0013	0.07	-0.0002	0.001	-0.0003	0.7467	1.3046							
	8	0.8270	0.0002	0.0037	0.1477	0.6553	-0.0013	-0.0002	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.6505	0.1446							
	9	0.0262	0.0007	0.0034	0.1477	0.6558	0.0019	-0.0002	0.0000	0.0000	0.0000	0.0000	0.0000	-0.1167	0.5266							
501	2	1.249	3.13	78.69	0.0418	0.01	-0.07	0.11	-0.0030	0.07	-0.0000	0.001	-0.0002	-0.1167	0.5266							
	8	0.8269	0.0002	0.0037	0.1448	0.6221	0.0030	-0.0000	0.0000	0.0000	-0.0000	0.0000	0.0000	-0.0520	0.6221							
	9	0.0268	0.0007	0.0035	0.1448	0.6621	0.0058	-0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0540	0.6460							
501	3	1.249	3.13	67.49	0.0434	0.01	-0.06	0.10	-0.0072	0.06	0.0000	0.001	-0.0009	0.0540	0.6460							
	8	0.8262	0.0002	0.0036	0.1465	0.6891	0.0103	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0350	0.6460							
	9	0.0259	0.0007	0.0033	0.1465	0.6500	0.0172	-0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0740	0.6588							
501	4	1.250	3.13	56.19	0.0462	0.01	-0.05	0.10	-0.0117	0.05	0.0001	0.001	-0.0015	0.0740	0.6588							
	8	0.8243	0.0002	0.0033	0.1642	0.6831	0.0149	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0504	0.6554							
	9	0.0232	0.0007	0.0030	0.1642	0.6505	0.0149	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0773	0.6605							
501	5	1.248	3.13	44.99	0.0481	0.00	-0.04	0.09	-0.0164	0.04	0.0002	0.002	-0.0025	0.0773	0.6605							
	8	0.8212	0.0002	0.0028	0.1883	0.6765	0.0193	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0588	0.6481							
	9	0.0193	0.0007	0.0024	0.1883	0.6336	0.0193	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0756	0.6609							
501	6	1.249	3.13	33.69	0.0427	-0.00	-0.04	0.09	-0.0207	0.04	0.0003	0.003	-0.0027	0.0756	0.6609							
	8	0.8171	0.0001	0.0023	0.2322	0.6774	0.0227	0.0003	0.0003	0.0003	0.0003	0.0003	0.0003	0.0629	0.6609							
	9	0.0144	0.0006	0.0018	0.2322	0.6291	0.0227	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0956	0.6799							
501	11	1.249	3.11	22.49	0.0300	-0.00	-0.03	0.08	-0.0232	0.03	0.0004	0.004	-0.0033	0.0956	0.6799							
	8	0.8124	0.0005	0.0017	0.2672	0.6886	0.0232	0.0004	0.0004	0.0004	0.0004	0.0004	0.0004	0.0593	0.6771							
	9	0.0093	0.0005	0.0011	0.2672	0.6074	0.0250	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0898	0.6763							
501	12	1.249	3.12	11.19	-0.0242	-0.01	-0.01	0.07	-0.0261	0.01	0.0004	0.004	-0.0035	0.0898	0.6763							
	8	0.8079	-0.0003	0.0010	-0.0480	0.6769	0.0265	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.1347	0.7134							
	9	0.0041	0.0003	0.0004	-0.0480	0.5909	0.0265	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0346	0.6687							
502	1	1.250	1.03	11.19	-0.0488	-0.01	-0.04	0.05	-0.0085	0.04	0.0002	0.002	-0.0012	0.1347	0.7134							
	8	0.8047	-0.0000	0.0006	1.2303	0.7352	0.0092	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0346	0.6687							
	9	0.0010	0.0002	0.0000	1.2303	0.1858	0.0092	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.1515	0.7116							
502	2	1.249	1.03	22.49	-0.0159	-0.01	-0.05	0.05	-0.0073	0.05	0.0002	0.002	-0.0010	0.1515	0.7116							
	8	0.8061	-0.0000	0.0008	-0.7574	0.6982	0.0073	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0233	0.6610							
	9	0.0021	0.0003	0.0002	-0.7574	0.5241	0.0089	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0233	0.6610							

See Key

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key											
502	3	1.250 0.0071 0.0033	-0.0000 0.0003	1.03	33.69 0.0010 0.0003	-0.0024 0.5377	-0.01 0.7375 0.5841	-0.05 0.0060 0.0080	0.06 0.0001 0.0000	-0.02 0.0008 0.0010	0.1275 0.0294	0.6884 0.6830	
502	4	1.249 0.0081 0.0047	-0.0000 0.0004	1.02	44.99 0.0011 0.0005	-0.0305 0.4283	-0.01 0.7068 0.6349	-0.06 0.0045 0.0069	0.07 0.0000 0.0000	-0.01 0.0006 0.0009	0.0916 0.0351	0.6688 0.6650	
502	5	1.249 0.0082 0.0058	-0.0000 0.0004	1.02	56.19 0.0011 0.0007	-0.0232 0.3861	-0.01 0.7245 0.6283	-0.06 0.0032 0.0056	0.07 0.0000 0.0000	-0.02 0.0003 0.0007	-0.0197 0.0029	0.6060 0.6606	
502	6	1.249 0.0082 0.0069	-0.0000 0.0004	1.01	67.49 0.0012 0.0008	-0.0156 0.3389	-0.01 0.7431 0.6444	-0.07 0.0014 0.0049	0.07 0.0000 0.0001	-0.02 0.0001 0.0005	-0.1949 0.1030	0.4128 0.5828	
502	7	1.250 0.0082 0.0077	0.0000 0.0004	1.02	78.69 0.0012 0.0009	0.0014 0.3062	-0.01 0.7551 0.6255	-0.06 0.0001 0.0037	0.07 0.0001 0.0001	-0.02 0.0001 0.0003	3.9893 0.2134	3.8136 0.4513	
502	8	1.250 0.0085 0.0079	-0.0000 0.0004	1.03	89.99 0.0012 0.0010	-0.0007 0.2970	-0.01 0.7303 0.6460	-0.07 0.0018 0.0020	0.07 0.0001 0.0001	-0.03 0.0003 0.0000	0.4839 0.4754	1.0255 0.2014	
503	1	1.249 0.0001 0.0006	-0.0001 0.0002	-0.02	89.99 0.0002 0.0000	-4.4867 -2.0307	-0.02 0.714 0.7328	-0.07 0.0012 0.0024	0.05 0.0001 0.0001	-0.02 0.0003 0.0001	0.7940 0.4049	1.3932 0.2753	
503	2	1.249 0.0002 0.0008	-0.0001 0.0002	-0.03	78.69 0.0002 0.0001	-3.2153 -1.6086	-0.02 0.711 0.8059	-0.07 0.0013 0.0024	0.05 0.0001 0.0001	-0.02 0.0003 0.0001	0.5730 0.3756	1.1947 0.3234	
503	3	1.249 0.0002 0.0008	-0.0001 0.0002	-0.03	67.49 0.0002 0.0001	-3.2823 -1.3936	-0.03 0.9158 1.0356	-0.07 0.0010 0.0024	0.05 0.0001 0.0001	-0.02 0.0002 0.0001	0.6782 0.3655	1.1892 0.4054	
503	4	1.250 0.0007 0.0010	-0.0001 0.0002	-0.02	56.19 0.0002 0.0001	-0.9281 -1.1011	-0.02 0.8069 0.8223	-0.06 0.0003 0.0024	0.05 0.0001 0.0001	-0.03 0.0001 0.0002	1.8715 0.3658	2.0038 0.4583	
503	5	1.250 0.0015 0.0012	-0.0001 0.0002	-0.03	44.99 0.0003 0.0001	-0.4046 -0.8948	-0.02 0.91909 0.7275	-0.06 0.0002 0.0025	0.05 0.0000 0.0001	-0.02 0.0000 0.0002	-1.8448 0.3361	-1.4396 0.4874	

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																	
503	6	7	1.249	-0.0021	0.0009	-0.003	0.0001	53.69	-0.2689	-1.1983	0.9357	-0.02	0.0009	-0.0025	0.05	0.0000	0.0000	-0.0002	-0.2215	0.3406
503	7	6	1.249	0.0025	-0.0007	-0.0000	0.0004	22.49	-0.1342	-1.5756	0.9214	-0.02	0.0014	-0.0024	0.05	0.0000	0.0000	-0.0001	-0.0633	0.5824
503	8	9	1.249	0.0028	-0.0003	-0.0002	0.0004	11.19	-0.1004	-3.0903	0.8573	-0.02	0.0017	-0.0023	0.06	0.0000	0.0000	-0.0001	-0.3186	0.5359
504	1	7	1.248	-0.0027	-0.0049	0.0000	-0.0002	11.19	0.4202	0.4873	0.4051	-0.03	-0.0260	-0.0227	0.04	0.0005	-0.0036	0.0964	0.6970	
504	2	6	1.248	0.0076	-0.0094	-0.0000	-0.0008	22.49	0.2321	0.0109	0.5694	-0.04	-0.0246	-0.0211	0.03	0.0004	-0.0033	0.1385	0.7012	
504	3	9	1.249	-0.0123	-0.0139	-0.0001	-0.0019	33.69	0.1791	0.0460	0.6073	-0.05	-0.0178	-0.0184	0.03	0.0004	-0.0030	0.0911	0.6854	
504	4	7	1.250	-0.0165	-0.0188	-0.0001	-0.0024	44.99	0.1515	0.0442	0.6224	-0.05	-0.0147	-0.0147	0.01	0.0004	-0.0021	0.1005	0.7142	
504	5	6	1.250	0.0204	-0.0226	-0.0001	-0.0030	56.19	0.1311	0.0436	0.6315	-0.06	-0.0136	-0.0103	0.02	0.0003	-0.0015	0.1452	0.7039	
504	6	9	1.249	-0.0234	-0.0256	-0.0002	-0.0034	67.49	0.1222	0.0482	0.6369	-0.06	-0.0093	-0.0060	0.01	0.0003	-0.0009	0.1826	0.7328	
504	7	7	1.249	-0.0252	-0.0276	-0.0002	-0.0036	78.69	0.1190	0.0466	0.6375	-0.07	-0.0047	-0.0016	0.00	0.0002	-0.0003	0.2699	0.8070	
504	8	6	1.249	-0.0259	-0.0280	-0.0002	-0.0037	89.99	0.1162	0.0515	0.6398	-0.07	-0.0020	-0.0001	0.00	0.0001	-0.0001	0.7507	1.0845	
504	9	9	1.249	-0.0280	-0.0300	-0.0002	-0.0037	99.99	0.1162	0.0515	0.6398	-0.07	-0.0020	-0.0001	0.00	0.0001	-0.0001	17.1210	14.6140	

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																				
505	1	7	1.249	-3.16	69.99	0.1382	0.97	-0.06	0.01	-0.02	-7.1157	-5.6772	0.0194	-0.0005	-0.0023	0.6128	0.0001	-0.0001	0.0001	-0.0001	-0.3449	0.3739	
		8	-0.0261	-0.0002	-0.0037	0.0499	0.6626	0.0020	-0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001
505	2	7	1.249	-3.17	76.69	0.1410	0.97	-0.07	0.00	-0.03	0.2795	0.8082	0.0168	-0.0005	-0.0022	0.6102	0.0002	-0.0002	0.0002	0.0002	0.6318	1.2017	
		8	-0.0276	-0.0002	-0.0036	0.0462	0.6579	-0.0015	-0.0002	-0.0003	0.0003	0.0003	0.0003	0.0003	0.0003	0.0003	0.0003	0.0003	0.0003	0.0003	0.0003	0.0003	0.0003
505	3	7	1.250	-3.17	67.49	0.1447	0.97	-0.09	0.01	-0.04	0.2002	0.7569	0.0170	-0.0004	-0.0020	0.6039	0.0003	-0.0003	0.0003	0.0003	0.2888	0.7931	
		8	-0.0258	-0.0002	-0.0034	0.0446	0.6634	-0.0058	-0.0003	-0.0009	0.0009	0.0009	0.0009	0.0009	0.0009	0.0009	0.0009	0.0009	0.0009	0.0009	0.0009	0.0009	0.0009
505	4	7	1.249	-3.16	56.19	0.1607	0.98	-0.09	0.01	-0.04	0.1519	0.7181	0.0141	-0.0004	-0.0016	0.5938	0.0004	-0.0004	0.0004	0.0004	0.2059	0.7631	
		8	-0.0226	-0.0001	-0.0029	0.0440	0.6621	-0.0102	-0.0004	-0.0015	0.0015	0.0015	0.0015	0.0015	0.0015	0.0015	0.0015	0.0015	0.0015	0.0015	0.0015	0.0015	0.0015
505	5	7	1.248	-3.16	44.99	0.1892	0.98	-0.10	0.02	-0.05	0.1238	0.7068	0.0106	-0.0004	-0.0024	0.5903	0.0004	-0.0004	0.0004	0.0004	0.1658	0.7220	
		8	-0.0187	-0.0001	-0.0024	0.0449	0.6677	-0.0146	-0.0004	-0.0021	0.0021	0.0021	0.0021	0.0021	0.0021	0.0021	0.0021	0.0021	0.0021	0.0021	0.0021	0.0021	0.0021
505	6	7	1.250	-3.16	33.69	0.2272	0.99	-0.10	0.02	-0.05	0.1060	0.6968	0.0067	-0.0003	-0.0007	0.5314	0.0004	-0.0004	0.0004	0.0004	0.1509	0.7133	
		8	-0.0140	-0.0001	-0.0018	0.0396	0.6676	-0.0183	-0.0005	-0.0026	0.0026	0.0026	0.0026	0.0026	0.0026	0.0026	0.0026	0.0026	0.0026	0.0026	0.0026	0.0026	0.0026
505	7	7	1.249	-3.17	22.49	0.5251	1.00	-0.11	0.04	-0.06	0.0970	0.6922	0.0019	-0.0002	-0.0013	0.2277	0.0004	-0.0004	0.0004	0.0004	0.1423	0.6982	
		8	-0.0095	-0.0000	-0.0013	0.0070	0.6898	-0.0212	-0.0006	-0.0029	0.0029	0.0029	0.0029	0.0029	0.0029	0.0029	0.0029	0.0029	0.0029	0.0029	0.0029	0.0029	0.0029
505	8	7	1.250	-3.17	11.19	-0.1388	1.01	-0.10	0.04	-0.06	0.0978	0.6926	0.0029	-0.0000	-0.0007	0.9434	0.0005	-0.0005	0.0005	0.0005	0.1375	0.6986	
		8	-0.0049	-0.0000	-0.0007	-0.0669	0.7591	-0.0229	-0.0006	-0.0032	0.0032	0.0032	0.0032	0.0032	0.0032	0.0032	0.0032	0.0032	0.0032	0.0032	0.0032	0.0032	0.0032
506	1	7	1.250	0.00	11.19	0.0360	1.02	-0.06	0.05	-0.03	-0.0537	0.6448	0.0086	0.0000	0.0012	0.7241	0.0000	-0.0000	0.0000	0.0000	-0.3189	0.5887	
		8	-0.0086	0.0000	0.0012	-3.3416	2.4352	0.0020	-0.0001	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002
506	2	7	1.250	0.00	22.49	0.0143	1.02	-0.06	0.05	-0.02	-0.1157	0.5324	0.0086	0.0000	0.0012	0.6987	0.0000	-0.0000	0.0000	0.0000	-0.2814	0.5807	
		8	-0.0086	0.0000	0.0012	-1.9381	1.6735	0.0025	-0.0001	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002
506	3	7	1.249	-0.00	33.69	0.0021	1.02	-0.06	0.05	-0.02	-0.2675	0.4146	0.0081	0.0000	0.0012	0.7497	0.0000	-0.0000	0.0000	0.0000	-0.2715	0.5472	
		8	-0.0081	0.0000	0.0012	-1.3838	0.8823	0.0024	-0.0001	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																
506	4	7	1.249	-0.0000	44.99	-0.0066	1.02	-0.006	0.05	-0.0000	-0.0000	-1.1168	-0.1070	0.0000	0.0000	-0.0002	-0.3350	0.4841
		8	0.0072	0.0000	0.0011	-0.0001	0.7918	0.0005	-0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001
		9	-0.0010	0.0002	-0.0001	-1.2199	0.7271	0.0026	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001
506	5	7	1.249	-0.0001	56.19	-0.0201	1.02	-0.006	0.05	-0.0001	-0.0001	-24.6500	-16.6732	0.0001	0.0001	0.0001	-0.3539	0.4569
		8	0.0068	0.0002	0.0010	-0.0001	0.7718	0.0000	-0.0001	-0.0001	-0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001
		9	-0.0009	0.0002	-0.0001	-1.3996	0.8149	0.0025	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001
506	6	7	1.249	-0.0000	67.49	-0.0290	1.02	-0.007	0.06	-0.0001	-0.0001	1.0955	0.4190	0.0000	0.0000	0.0001	-0.3760	0.4035
		8	0.0063	0.0000	0.0009	-0.0001	0.7887	0.0007	-0.0001	-0.0001	-0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001
		9	-0.0008	0.0002	-0.0001	-1.7056	0.9223	0.0024	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001
506	7	7	1.249	-0.0000	78.69	-0.0288	1.02	-0.012	0.06	-0.0001	-0.0001	0.7245	1.3037	0.0000	0.0000	0.0001	-0.3891	0.3327
		8	0.0063	0.0000	0.0010	-0.0001	0.7942	0.0012	-0.0001	-0.0001	-0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001
		9	-0.0006	0.0002	-0.0001	-2.2653	0.9802	0.0023	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001
506	8	7	1.250	-0.0000	89.99	-0.0281	1.02	-0.011	0.05	-0.0001	-0.0001	0.9423	1.3970	0.0000	0.0000	0.0001	-0.4444	0.3036
		8	0.0061	0.0000	0.0009	-0.0001	0.8052	0.0011	-0.0001	-0.0001	-0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001
		9	-0.0003	0.0002	-0.0001	-4.4596	0.9465	0.0022	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001
507	1	7	1.249	1.0000	89.99	0.0297	1.03	-0.014	0.07	0.0001	0.0001	0.6858	1.2102	0.0000	0.0000	0.0001	-0.4899	0.2740
		8	0.0146	0.0000	0.0020	0.0000	0.7021	0.0014	-0.0001	-0.0001	-0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001
		9	0.0077	0.0004	0.0009	0.3215	0.6477	0.0020	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001
507	2	7	1.250	1.0000	78.69	0.0306	1.03	-0.006	0.07	0.0001	0.0001	-9.9386	-0.0337	0.0000	0.0000	0.0001	-0.2273	0.4885
		8	0.0141	0.0000	0.0020	0.0000	0.7121	0.0000	-0.0001	-0.0001	-0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001
		9	0.0074	0.0005	0.0009	0.3451	0.6518	0.0035	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001
507	3	7	1.250	1.0000	67.49	0.0243	1.03	-0.015	0.07	0.0001	0.0001	-0.2622	0.3802	0.0000	0.0000	0.0001	-0.1070	0.5797
		8	0.0142	0.0000	0.0019	0.0000	0.7010	0.0015	-0.0001	-0.0001	-0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001
		9	0.0066	0.0005	0.0008	0.3834	0.6510	0.0048	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001
507	4	7	1.250	1.0000	56.19	0.0220	1.02	-0.006	0.06	0.0000	0.0000	-0.0777	0.5888	0.0000	0.0000	0.0001	-0.0009	0.6645
		8	0.0141	0.0000	0.0019	0.0000	0.7056	0.0033	-0.0000	-0.0000	-0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
		9	0.0055	0.0004	0.0007	0.4428	0.6500	0.0056	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000
507	5	7	1.249	1.0000	44.99	0.0212	1.03	-0.005	0.06	0.0000	0.0000	0.0575	0.6520	0.0000	0.0000	0.0001	0.0314	0.6660
		8	0.0140	0.0000	0.0019	0.0000	0.6986	0.0046	-0.0000	-0.0000	-0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
		9	0.0043	0.0004	0.0005	0.5105	0.6565	0.0068	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000
507	6	7	1.250	1.0000	33.69	0.0299	1.03	-0.005	0.06	0.0000	0.0000	0.1062	0.6860	0.0000	0.0000	0.0001	0.0310	0.6785
		8	0.0132	0.0000	0.0018	0.0000	0.6975	0.0062	-0.0000	-0.0000	-0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
		9	0.0030	0.0004	0.0003	0.6538	0.6382	0.0080	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																		
507	7	1.250	1.00	22.49	0.0399	1.03	-0.05	0.06	-0.01	0.1235	0.6981									
	8	0.0119	0.0000	0.0016	0.8936	0.6909	0.0076	0.0001	0.0010	0.0257	0.6813									
	9	0.0020	0.0003	0.0002	0.0000	0.5421	0.0086	0.0000	0.0011											
507	7	1.249	1.00	11.19	0.0287	1.02	-0.05	0.05	-0.01	0.1142	0.7039									
	8	0.0104	0.0000	0.0014	2.2630	0.6961	0.0088	0.0002	0.0012	0.0359	0.6748									
	9	0.0006	0.0002	0.0000	0.0000	0.3595	0.0089	0.0000	0.0012											
508	7	1.249	3.11	11.20	0.0221	1.02	-0.01	0.06	0.00	0.0899	0.6698									
	8	0.0135	0.0000	0.0018	0.4799	0.6840	0.0266	0.0004	0.0035	0.0510	0.6755									
	9	0.0040	0.0003	0.0004	0.0000	0.6118	0.0266	0.0002	0.0036											
508	7	1.249	3.11	22.49	0.0379	1.03	-0.02	0.07	0.00	0.0897	0.6691									
	8	0.0185	0.0001	0.0025	0.2946	0.6836	0.0236	0.0004	0.0031	0.0556	0.6643									
	9	0.0091	0.0005	0.0011	0.0000	0.6456	0.0252	0.0002	0.0033											
508	7	1.248	3.11	33.69	0.0516	1.04	-0.02	0.09	-0.00	0.0879	0.6704									
	8	0.0230	0.0012	0.0031	0.2200	0.6920	0.0200	0.0003	0.0026	0.0636	0.6634									
	9	0.0144	0.0006	0.0018	0.0000	0.6369	0.0226	0.0002	0.0030											
508	7	1.275	3.11	44.99	0.0529	1.04	-0.03	0.09	-0.00	0.0910	0.6726									
	8	0.0193	0.0007	0.0025	0.1815	0.6877	0.0157	0.0002	0.0021	0.0596	0.6561									
	9	0.0000	0.0000	0.0000	0.0000	0.6484	0.0192	0.0002	0.0025											
508	7	1.248	3.12	56.19	0.0493	1.05	-0.04	0.09	-0.01	0.0805	0.6540									
	8	0.0307	0.0003	0.0042	0.1610	0.6923	0.0115	0.0001	0.0015	0.0538	0.6586									
	9	0.0231	0.0007	0.0030	0.0000	0.6637	0.0149	0.0001	0.0019											
508	7	1.249	3.12	67.49	0.0457	1.05	-0.05	0.10	-0.01	0.0710	0.6410									
	8	0.0325	0.0002	0.0045	0.1417	0.6923	0.0068	0.0000	0.0008	0.0291	0.6361									
	9	0.0259	0.0007	0.0034	0.0000	0.6672	0.0104	0.0000	0.0013											
508	7	1.250	3.12	78.69	0.0434	1.05	-0.06	0.10	-0.02	-0.0937	0.5389									
	8	0.0333	0.0002	0.0046	0.1382	0.6952	0.0027	-0.0000	0.0007	-0.0487	0.6367									
	9	0.0268	0.0007	0.0035	0.0000	0.6644	0.0057	-0.0000	0.0007											
508	7	1.250	3.12	89.99	0.0422	1.05	-0.07	0.10	-0.03	0.5765	1.1673									
	8	0.0333	0.0002	0.0046	0.1414	0.6958	-0.0016	-0.0001	-0.0003	-0.6371	0.1745									
	9	0.0260	0.0007	0.0034	0.0000	0.6654	-0.0017	-0.0002	-0.0000											
509	7	1.249	6.31	89.99	0.0408	1.09	-0.06	0.14	-0.03	0.6830	1.3247									
	8	0.0637	0.0005	0.0085	0.0848	0.6711	-0.0010	-0.0001	-0.0002	-1.9496	-0.3146									
	9	0.0546	0.0009	0.0073	0.0000	0.6689	-0.0006	-0.0002	-0.0000											

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key													
509	2	7	1.249	6.31	78.69	0.432	1.09	-0.04	0.14	-0.01	0.0895	0.6373				
		8	0.0622	0.0005	0.0083	0.0432	0.6732	0.0078	0.0001	0.0010	0.0271	0.6558				
		9	0.0546	0.0009	0.0073	0.0846	0.6697	0.0096	0.0000	0.0012						
509	3	7	1.249	6.29	67.49	0.451	1.09	-0.03	0.14	-0.00	0.0789	0.6512				
		8	0.0589	0.0005	0.0079	0.0451	0.6715	0.0181	0.0002	0.0023	0.0602	0.6592				
		9	0.0526	0.0009	0.0070	0.0869	0.6691	0.0203	0.0000	0.0026						
509	4	7	1.249	6.30	56.19	0.452	1.08	-0.01	0.14	0.00	0.0778	0.6620				
		8	0.0538	0.0004	0.0072	0.0452	0.6759	0.0277	0.0004	0.0036	0.0533	0.6639				
		9	0.0486	0.0009	0.0065	0.0921	0.6698	0.0308	0.0003	0.0041						
509	5	7	1.249	6.30	44.99	0.433	1.07	-0.00	0.13	0.02	0.0724	0.6691				
		8	0.0468	0.0004	0.0063	0.0433	0.6754	0.0367	0.0005	0.0049	0.0481	0.6710				
		9	0.0422	0.0008	0.0056	0.0994	0.6732	0.0402	0.0003	0.0054						
509	6	7	1.248	6.29	33.70	0.457	1.06	0.00	0.11	0.01	0.0678	0.6705				
		8	0.0379	0.0003	0.0051	0.0457	0.6766	0.0444	0.0006	0.0059	0.0416	0.6747				
		9	0.0325	0.0008	0.0043	0.1251	0.6749	0.0480	0.0003	0.0064						
509	7	7	1.249	6.30	22.50	0.522	1.04	0.01	0.10	0.01	0.0670	0.6729				
		8	0.0278	0.0002	0.0037	0.0522	0.6760	0.0501	0.0006	0.0067	0.0376	0.6699				
		9	0.0218	0.0007	0.0028	0.1690	0.6570	0.0536	0.0004	0.0071						
509	8	7	1.248	6.30	11.20	0.297	1.03	0.02	0.07	0.03	0.0662	0.6702				
		8	0.0174	0.0001	0.0023	0.0297	0.6671	0.0542	0.0007	0.0072	0.0370	0.6703				
		9	0.0098	0.0006	0.0013	0.3031	0.6650	0.0571	0.0004	0.0076						
510	1	7	1.250	9.48	11.20	0.327	1.03	0.05	0.08	0.05	0.0628	0.6508				
		8	0.0221	0.0001	0.0029	0.0327	0.6718	0.0790	0.0009	0.0102	0.0381	0.6488				
		9	0.0150	0.0006	0.0019	0.2060	0.6590	0.0847	0.0006	0.0110						
510	2	7	1.249	9.48	22.50	0.380	1.06	0.05	0.11	0.05	0.0626	0.6551				
		8	0.0379	0.0002	0.0051	0.0380	0.6815	0.0736	0.0009	0.0096	0.0390	0.6530				
		9	0.0340	0.0007	0.0046	0.1096	0.6822	0.0812	0.0006	0.0106						
510	3	7	1.249	9.47	33.70	0.389	1.08	0.04	0.14	0.04	0.0647	0.6610				
		8	0.0523	0.0004	0.0070	0.0389	0.6742	0.0657	0.0008	0.0086	0.0392	0.6523				
		9	0.0503	0.0009	0.0067	0.0947	0.6740	0.0749	0.0005	0.0097						
510	4	7	1.250	9.47	45.00	0.418	1.10	0.02	0.16	0.04	0.0645	0.6644				
		8	0.0647	0.0005	0.0086	0.0418	0.6672	0.0552	0.0007	0.0073	0.0397	0.6631				
		9	0.0632	0.0011	0.0083	0.0891	0.6606	0.0648	0.0005	0.0085						

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key →																			
510	5	7	1.248	9.47	56.13	0.0420	1.11	-0.00	0.18	0.02	0.0664	0.6668	0.0718	0.0006	0.0094	0.0420	0.6597	0.0516	0.0003	0.0056	0.0356	0.6622
510	6	9	0.0718	0.0012	0.0094	0.0874	0.6568	0.0423	0.0005	0.0068	0.0664	0.6668	0.0718	0.0006	0.0094	0.0420	0.6597	0.0516	0.0003	0.0056	0.0356	0.6622
510	7	7	1.248	9.49	67.49	0.0421	1.12	-0.01	0.19	0.00	0.0712	0.6576	0.0777	0.0012	0.0100	0.0421	0.6482	0.0341	0.0002	0.0045	0.0358	0.6606
510	8	9	0.0777	0.0012	0.0100	0.0834	0.6482	0.0276	0.0003	0.0036	0.0712	0.6576	0.0777	0.0012	0.0100	0.0421	0.6482	0.0341	0.0002	0.0045	0.0358	0.6606
510	7	7	1.248	9.49	78.69	0.0432	1.13	-0.04	0.19	-0.01	0.0754	0.6351	0.0868	0.0013	0.0104	0.0432	0.6455	0.0157	0.0001	0.0020	0.0434	0.6513
510	8	9	0.0868	0.0013	0.0104	0.0816	0.6455	0.0125	0.0001	0.0015	0.0754	0.6351	0.0868	0.0013	0.0104	0.0432	0.6455	0.0157	0.0001	0.0020	0.0434	0.6513
510	7	7	1.249	9.50	89.99	0.0453	1.13	-0.06	0.19	-0.03	0.6551	1.3523	0.0891	0.0012	0.0103	0.0453	0.6484	0.0002	-0.0001	-0.0002	0.6551	1.3523
510	8	9	0.0891	0.0012	0.0103	0.0794	0.6484	-0.0010	0.0001	-0.0002	0.6551	1.3523	0.0891	0.0012	0.0103	0.0453	0.6484	0.0002	-0.0001	-0.0002	0.6551	1.3523
511	1	7	1.250	12.66	89.99	0.0460	1.16	-0.06	0.22	-0.04	0.7304	1.3422	1.091	0.0014	0.0125	0.0460	0.6253	0.0006	-0.0002	-0.0000	0.7304	1.3422
511	8	9	0.1091	0.0014	0.0125	0.0742	0.6253	-0.0010	0.0001	-0.0002	0.7304	1.3422	1.091	0.0014	0.0125	0.0460	0.6253	0.0006	-0.0002	-0.0000	0.7304	1.3422
511	2	7	1.250	12.66	78.69	0.0430	1.16	-0.03	0.22	-0.01	0.0801	0.6495	1.062	0.0015	0.0125	0.0430	0.6197	0.0226	0.0001	0.0030	0.0354	0.6700
511	8	9	0.1062	0.0015	0.0125	0.0751	0.6197	0.0184	0.0002	0.0001	0.0801	0.6495	1.062	0.0015	0.0125	0.0430	0.6197	0.0226	0.0001	0.0030	0.0354	0.6700
511	3	7	1.250	12.65	67.48	0.0416	1.15	-0.00	0.22	0.02	0.0741	0.6710	1.0985	0.0015	0.0116	0.0416	0.6220	0.0483	0.0003	0.0064	0.0361	0.6678
511	8	9	0.0985	0.0015	0.0116	0.0766	0.6220	0.0387	0.0005	0.0051	0.0741	0.6710	1.0985	0.0015	0.0116	0.0416	0.6220	0.0483	0.0003	0.0064	0.0361	0.6678
511	4	7	1.247	12.64	56.18	0.0431	1.14	0.02	0.21	0.04	0.0699	0.6668	1.0919	0.0014	0.0116	0.0431	0.6402	0.0572	0.0005	0.0091	0.0429	0.6593
511	8	9	0.0927	0.0014	0.0116	0.0786	0.6296	0.0690	0.0005	0.0091	0.0429	0.6593	1.0919	0.0014	0.0116	0.0431	0.6402	0.0572	0.0005	0.0091	0.0429	0.6593
511	5	7	1.248	12.63	45.00	0.0434	1.12	0.04	0.19	0.05	0.0647	0.6558	1.0804	0.0014	0.0105	0.0434	0.6510	0.0722	0.0009	0.0094	0.0427	0.6475
511	8	9	0.0818	0.0014	0.0105	0.0862	0.6423	0.0690	0.0005	0.0094	0.0427	0.6475	1.0804	0.0014	0.0105	0.0434	0.6510	0.0722	0.0009	0.0094	0.0427	0.6475
511	6	7	1.249	12.64	56.18	0.0435	1.14	0.02	0.21	0.04	0.0685	0.6645	1.0917	0.0014	0.0116	0.0435	0.6415	0.0573	0.0005	0.0091	0.0425	0.6590
511	8	9	0.0924	0.0014	0.0116	0.0797	0.6319	0.0690	0.0005	0.0091	0.0425	0.6590	1.0917	0.0014	0.0116	0.0435	0.6415	0.0573	0.0005	0.0091	0.0425	0.6590
511	7	7	1.249	12.65	67.48	0.0425	1.15	-0.00	0.22	0.02	0.0722	0.6627	1.0002	0.0015	0.0122	0.0425	0.6315	0.0391	0.0003	0.0051	0.0374	0.6627
511	8	9	0.0984	0.0015	0.0122	0.0776	0.6219	0.0484	0.0003	0.0051	0.0374	0.6627	1.0002	0.0015	0.0122	0.0425	0.6315	0.0391	0.0003	0.0051	0.0374	0.6627

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																	
511	8	1.248	12.66	78.69	0.0435	1.16	-0.03	0.22	-0.01	0.0764	0.6379	0.1063	0.0009	0.0133	0.0002	0.0001	0.0023	0.0764	0.6379
		0.1013	0.0015	0.0125	0.0761	0.6217	0.0230	0.0001	0.0030	0.0305	0.6622	0.01013	0.00015	0.0125	0.0002	0.0001	0.0030	0.0305	0.6622
511	9	1.249	12.66	89.99	0.0458	1.255	-0.007	0.22	-0.03	1.0594	1.7665	0.1090	0.0010	0.0136	-0.0002	-0.0000	-0.0000	1.0594	1.7665
		0.1009	0.0014	0.0125	0.0738	0.6197	0.0007	-0.0002	-0.0000	-1.3763	-0.1227	0.1009	0.0014	0.0125	-0.0002	-0.0000	-0.0000	-1.3763	-0.1227
512	1	1.249	12.67	89.99	0.0441	3.18	-0.06	0.22	-0.04	0.6961	1.3672	0.1168	0.0010	0.0144	-0.0001	-0.0000	-0.0003	0.6961	1.3672
		0.1010	0.0014	0.0125	0.0736	0.6199	0.0007	-0.0001	-0.0000	-1.2599	0.0829	0.1010	0.0014	0.0125	-0.0001	-0.0000	-0.0003	-1.2599	0.0829
512	2	1.249	12.66	78.69	0.0421	3.18	-0.03	0.22	-0.00	0.0867	0.6439	0.1137	0.0009	0.0141	0.0003	0.0003	0.0023	0.0867	0.6439
		0.1015	0.0015	0.0125	0.0753	0.6183	0.0229	0.0001	0.0030	0.0327	0.6685	0.1015	0.0015	0.0125	0.0003	0.0003	0.0023	0.0327	0.6685
512	3	1.250	12.66	67.48	0.0410	3.17	0.00	0.22	0.02	0.0750	0.6686	1.083	0.0008	0.0135	0.0003	0.0064	0.051	0.0750	0.6686
		0.0985	0.0015	0.0122	0.0777	0.6226	0.0483	0.0003	0.0064	0.0366	0.6678	0.0985	0.0015	0.0122	0.0003	0.0064	0.051	0.0366	0.6678
512	4	1.249	12.64	56.19	0.0409	3.16	0.02	0.21	0.03	0.0685	0.6579	1.005	0.0014	0.0127	0.0005	0.0090	0.076	0.0685	0.6579
		0.0928	0.0014	0.0116	0.0789	0.6292	0.0689	0.0005	0.0090	0.0418	0.6579	0.0928	0.0014	0.0116	0.0005	0.0090	0.076	0.0418	0.6579
512	5	1.248	12.65	45.00	0.0403	3.14	0.05	0.20	0.05	0.0635	0.6550	0.894	0.0014	0.0115	0.0007	0.0110	0.094	0.0635	0.6550
		0.0821	0.0014	0.0105	0.0853	0.6393	0.0848	0.0007	0.0110	0.0436	0.6490	0.0821	0.0014	0.0105	0.0007	0.0110	0.094	0.0436	0.6490
512	6	1.249	12.65	33.71	0.0401	3.12	0.06	0.17	0.06	0.0594	0.6442	0.751	0.0012	0.0087	0.0008	0.0122	0.108	0.0594	0.6442
		0.0669	0.0012	0.0087	0.0932	0.6515	0.0960	0.0008	0.0122	0.0436	0.6386	0.0669	0.0012	0.0087	0.0008	0.0122	0.108	0.0436	0.6386
512	7	1.249	12.61	22.51	0.0380	3.10	0.07	0.14	0.07	0.0586	0.6377	0.580	0.0004	0.0078	0.0008	0.0131	0.118	0.0586	0.6377
		0.0472	0.0010	0.0063	0.1060	0.6696	0.1036	0.0008	0.0131	0.0421	0.6350	0.0472	0.0010	0.0063	0.0008	0.0131	0.118	0.0421	0.6350
512	8	1.248	12.64	11.20	0.0352	3.07	0.08	0.10	0.08	0.0595	0.6314	0.376	0.0002	0.0051	0.0008	0.0133	0.125	0.0595	0.6314
		0.0217	0.0006	0.0028	0.1516	0.6512	0.1063	0.0008	0.0133	0.0413	0.6269	0.0217	0.0006	0.0028	0.0008	0.0133	0.125	0.0413	0.6269
513	4	1.249	9.44	11.20	0.0364	3.07	0.04	0.09	0.05	0.0656	0.6547	0.332	0.0002	0.0046	0.0010	0.0109	0.102	0.0656	0.6547
		0.0155	0.0005	0.0020	0.1822	0.6566	0.0847	0.0006	0.0109	0.0371	0.6467	0.0155	0.0005	0.0020	0.0006	0.0109	0.102	0.0371	0.6467

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key													
513	5	1.249 0.0492 0.0345	9.44 0.0003 0.0007	22.51 0.0067 0.0047	0.0360 0.1022	3.09 0.6880 0.6843	0.03 0.0728 0.0808	0.11 0.0009 0.0006	0.04 0.0095 0.0105	0.0643 0.0376	0.6562 0.6509				
513	6	1.249 0.0630 0.0503	9.43 0.0004 0.0009	33.71 0.0084 0.0068	0.0372 0.0927	3.11 0.728 0.6776	0.02 0.0653 0.0746	0.15 0.0008 0.0005	0.05 0.0086 0.0097	0.0649 0.0389	0.6597 0.6522				
513	7	1.250 0.0747 0.0634	9.44 0.0006 0.0010	45.00 0.0099 0.0083	0.0408 0.0865	3.13 0.644 0.6617	0.01 0.0344 0.0642	0.16 0.0007 0.0005	0.04 0.0072 0.0085	0.0688 0.0397	0.6673 0.6628				
513	8	1.249 0.0640 0.0718	9.44 0.0006 0.0012	56.19 0.0109 0.0094	0.0410 0.0847	3.14 0.534 0.6567	-0.00 0.0417 0.0513	0.18 0.0005 0.0003	0.02 0.0055 0.0067	0.0701 0.0341	0.6687 0.6619				
513	9	1.248 0.0908 0.0775	9.43 0.0007 0.0012	67.49 0.0117 0.0100	0.0424 0.0815	3.16 0.472 0.6507	-0.03 0.0269 0.0340	0.18 0.0004 0.0002	0.00 0.0035 0.0044	0.0779 0.0332	0.6620 0.6576				
513	10	1.249 0.0958 0.0803	9.45 0.0008 0.0012	78.69 0.0122 0.0103	0.0420 0.0795	3.16 0.402 0.6466	-0.05 0.0121 0.0158	0.19 0.0002 0.0001	-0.01 0.0015 0.0020	0.0860 0.0411	0.6281 0.6461				
513	11	1.248 0.0981 0.0803	9.45 0.0008 0.0012	89.99 0.0125 0.0103	0.0439 0.0777	3.17 0.392 0.6443	-0.07 -0.0016 -0.0002	0.19 -0.0001 -0.0001	-0.03 -0.0003 -0.0001	0.3002 -2.7063	1.1067 -2.6831				
514	1	1.250 0.0745 0.0547	6.28 0.0006 0.0008	89.99 0.0099 0.0072	0.0430 0.0817	3.13 0.6642 0.6613	-0.07 -0.0011 -0.0004	0.15 -0.0001 -0.0002	-0.03 -0.0002 -0.0000	0.4479 -2.8947	1.1747 -1.1544				
514	2	1.250 0.0731 0.0544	6.27 0.0006 0.0009	78.69 0.0097 0.0072	0.0430 0.0828	3.12 0.6658 0.6651	-0.05 0.0076 0.0097	0.15 0.0001 0.0000	-0.03 0.0009 0.0012	0.1081 0.0257	0.6474 0.6461				
514	3	1.250 0.0697 0.0524	6.27 0.0005 0.0008	67.49 0.0092 0.0070	0.0426 0.0857	3.12 0.654 0.6690	-0.04 0.0176 0.0202	0.15 0.0003 0.0002	-0.01 0.0023 0.0026	0.0866 0.0592	0.6527 0.6593				
514	4	1.249 0.0647 0.0486	6.27 0.0005 0.0008	56.19 0.0086 0.0064	0.0416 0.0910	3.10 0.6677 0.6673	-0.02 0.0274 0.0308	0.14 0.0004 0.0003	-0.00 0.0036 0.0040	0.0834 0.0517	0.6619 0.6602				

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																			
514	5	7	1.249	6.26	44.99	0.0414	3.10	-0.01	0.0005	0.013	0.0046	0.0763	0.6707	0.0423	0.0980	0.6752	0.0401	0.0003	0.0053	0.0471	0.6672	
		6	0.0582	0.0004	0.0078	0.00980	0.6752	0.0362	0.0005	0.0005	0.0046	0.0471	0.6672	0.0423	0.0980	0.6752	0.0401	0.0003	0.0053	0.0471	0.6672	
		9	0.0423	0.0008	0.0056	0.0980	0.6752	0.0362	0.0005	0.0005	0.0046	0.0471	0.6672	0.0423	0.0980	0.6752	0.0401	0.0003	0.0053	0.0471	0.6672	
514	6	7	1.249	6.26	33.70	0.0416	3.09	-0.00	0.0006	0.12	0.0058	0.0691	0.6713	0.0328	0.1202	0.6719	0.0476	0.0004	0.0064	0.0421	0.6717	
		8	0.0495	0.0004	0.0067	0.0416	0.6808	0.0438	0.0006	0.0006	0.0006	0.0058	0.0691	0.6713	0.0328	0.1202	0.6719	0.0476	0.0004	0.0064	0.0421	0.6717
		9	0.0328	0.0007	0.0044	0.1202	0.6719	0.0476	0.0004	0.0004	0.0004	0.0058	0.0691	0.6713	0.0328	0.1202	0.6719	0.0476	0.0004	0.0064	0.0421	0.6717
514	7	7	1.250	6.26	22.50	0.0454	3.08	0.00	0.0006	0.11	0.0066	0.0692	0.6722	0.0221	0.1664	0.6586	0.0495	0.0003	0.0071	0.0568	0.6689	
		8	0.0396	0.0003	0.0054	0.0454	0.6848	0.0495	0.0006	0.0006	0.0006	0.0066	0.0692	0.6722	0.0221	0.1664	0.6586	0.0495	0.0003	0.0071	0.0568	0.6689
		9	0.0221	0.0007	0.0029	0.1664	0.6586	0.0495	0.0003	0.0003	0.0003	0.0066	0.0692	0.6722	0.0221	0.1664	0.6586	0.0495	0.0003	0.0071	0.0568	0.6689
514	8	7	1.242	6.26	11.20	0.0363	3.06	0.00	0.0007	0.08	0.0071	0.0670	0.6689	0.0103	0.2808	0.6271	0.0537	0.0004	0.0075	0.0576	0.6706	
		8	0.0287	0.0002	0.0039	0.0363	0.6818	0.0537	0.0007	0.0007	0.0007	0.0071	0.0670	0.6689	0.0103	0.2808	0.6271	0.0537	0.0004	0.0075	0.0576	0.6706
		9	0.0103	0.0005	0.0013	0.2808	0.6271	0.0537	0.0004	0.0004	0.0004	0.0071	0.0670	0.6689	0.0103	0.2808	0.6271	0.0537	0.0004	0.0075	0.0576	0.6706
515	1	7	1.249	3.12	11.20	0.0380	3.05	-0.03	0.0004	0.07	0.0035	0.0646	0.6605	0.0252	0.5134	0.5981	0.0267	0.0002	0.0035	0.0510	0.6670	
		8	0.0252	0.0001	0.0034	0.0380	0.6817	0.0267	0.0004	0.0004	0.0035	0.0646	0.6605	0.6670	0.0252	0.5134	0.5981	0.0267	0.0002	0.0035	0.0510	0.6670
		9	0.0041	0.0004	0.0005	0.5134	0.5981	0.0266	0.0002	0.0002	0.0035	0.0646	0.6605	0.6670	0.0041	0.3083	0.6436	0.0251	0.0002	0.0033	0.0557	0.6576
515	2	7	1.249	3.11	22.50	0.0411	3.06	-0.03	0.0003	0.08	0.0031	0.0633	0.6577	0.0092	0.3083	0.6436	0.0251	0.0003	0.0033	0.0557	0.6576	
		8	0.0306	0.0002	0.0041	0.0411	0.6825	0.0239	0.0003	0.0003	0.0031	0.0633	0.6577	0.6576	0.0092	0.3083	0.6436	0.0251	0.0003	0.0033	0.0557	0.6576
		9	0.0092	0.0005	0.0011	0.3083	0.6436	0.0251	0.0002	0.0002	0.0033	0.0633	0.6577	0.6576	0.0092	0.3083	0.6436	0.0251	0.0003	0.0033	0.0557	0.6576
515	3	7	1.250	3.12	33.69	0.0478	3.07	-0.03	0.0003	0.09	0.0026	0.0607	0.6609	0.0354	0.6401	0.6401	0.0201	0.0002	0.0029	0.0627	0.6577	
		8	0.0354	0.0003	0.0049	0.0478	0.6929	0.0201	0.0003	0.0003	0.0026	0.0607	0.6609	0.6577	0.0354	0.6401	0.6401	0.0201	0.0002	0.0029	0.0627	0.6577
		9	0.0144	0.0006	0.0018	0.6401	0.6401	0.0225	0.0002	0.0002	0.0029	0.0627	0.6577	0.6576	0.0144	0.2326	0.6401	0.0225	0.0002	0.0029	0.0627	0.6577
515	4	7	1.249	3.12	44.99	0.0482	3.08	-0.05	0.0002	0.10	0.0020	0.0604	0.6604	0.0398	0.6480	0.6480	0.0156	0.0002	0.0024	0.0544	0.6475	
		8	0.0398	0.0003	0.0055	0.0482	0.6925	0.0156	0.0002	0.0002	0.0020	0.0604	0.6604	0.6475	0.0398	0.6480	0.6480	0.0156	0.0002	0.0024	0.0544	0.6475
		9	0.0196	0.0007	0.0025	0.6480	0.6480	0.0191	0.0002	0.0002	0.0024	0.0604	0.6604	0.6475	0.0196	0.6480	0.6480	0.0191	0.0002	0.0024	0.0544	0.6475
515	5	7	1.250	3.12	56.19	0.0455	3.09	-0.05	0.0001	0.10	0.0014	0.0623	0.6580	0.0431	0.6610	0.6610	0.0112	0.0001	0.0019	0.0488	0.6514	
		8	0.0431	0.0003	0.0059	0.0455	0.6918	0.0112	0.0001	0.0001	0.0014	0.0623	0.6580	0.6514	0.0431	0.6610	0.6610	0.0112	0.0001	0.0019	0.0488	0.6514
		9	0.0234	0.0007	0.0030	0.6610	0.6610	0.0149	0.0001	0.0001	0.0019	0.0623	0.6580	0.6514	0.0234	0.6610	0.6610	0.0149	0.0001	0.0019	0.0488	0.6514
515	6	7	1.251	3.13	67.49	0.0442	3.09	-0.06	0.0000	0.11	0.0008	0.0582	0.6299	0.0449	0.6685	0.6685	0.0068	0.0000	0.0013	0.0288	0.6506	
		8	0.0449	0.0003	0.0062	0.0442	0.6913	0.0068	0.0000	0.0000	0.0008	0.0582	0.6299	0.6506	0.0449	0.6685	0.6685	0.0068	0.0000	0.0013	0.0288	0.6506
		9	0.0260	0.0007	0.0034	0.6685	0.6685	0.0101	0.0000	0.0000	0.0013	0.0288	0.6506	0.6506	0.0260	0.6685	0.6685	0.0101	0.0000	0.0013	0.0288	0.6506
515	7	7	1.251	3.13	78.69	0.0450	3.09	-0.07	0.0000	0.10	0.0003	0.0937	0.5262	0.0456	0.6582	0.6582	0.0028	0.0000	0.0007	0.0482	0.6349	
		8	0.0456	0.0004	0.0063	0.0450	0.6958	0.0028	0.0000	0.0000	0.0003	0.0937	0.5262	0.6349	0.0456	0.6582	0.6582	0.0028	0.0000	0.0007	0.0482	0.6349
		9	0.0270	0.0007	0.0035	0.6582	0.6582	0.0056	0.0000	0.0000	0.0007	0.0482	0.5262	0.6349	0.0270	0.6582	0.6582	0.0056	0.0000	0.0007	0.0482	0.6349

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																
515	7	1.249	0.0455	0.0455	0.6968	0.6613	0.0014	0.0018	0.0014	0.0011	0.0001	0.0003	0.0000	0.0003	0.6257	0.6181	1.2257	0.1457
	8	0.0456	0.0004	0.0007	0.0063	0.0034	0.0007	0.0007	0.0007	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001
	9	0.0263	0.0007	0.0007	0.0034	0.0034	0.0007	0.0007	0.0007	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001
516	1	1.249	0.0458	0.0458	0.7039	0.6777	0.0013	0.0020	0.0013	0.0001	0.0001	0.0003	0.0001	0.0003	0.7059	0.4757	1.3907	0.2475
	7	0.0267	0.0002	0.0005	0.0037	0.0010	0.0002	0.0002	0.0002	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001
	8	0.0075	0.0005	0.0005	0.0010	0.0010	0.0005	0.0005	0.0005	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001
516	2	1.249	0.0453	0.0428	0.7096	0.6629	0.0006	0.0036	0.0006	0.0001	0.0001	0.0003	0.0001	0.0003	0.7057	0.2173	0.5548	0.4520
	7	0.0264	0.0002	0.0005	0.0037	0.0009	0.0002	0.0002	0.0002	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001
	8	0.0074	0.0005	0.0005	0.0009	0.0009	0.0005	0.0005	0.0005	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001
516	3	1.249	0.0412	0.0364	0.7041	0.6241	0.0014	0.0048	0.0014	0.0000	0.0000	0.0001	0.0000	0.0001	0.2465	0.1035	0.3622	0.5796
	7	0.0261	0.0002	0.0005	0.0036	0.0008	0.0002	0.0002	0.0002	0.0000	0.0000	0.0001	0.0000	0.0001	0.0001	0.0001	0.0001	0.0001
	8	0.0068	0.0005	0.0005	0.0008	0.0008	0.0005	0.0005	0.0005	0.0000	0.0000	0.0001	0.0000	0.0001	0.0001	0.0001	0.0001	0.0001
516	4	1.249	0.0404	0.0426	0.7024	0.6502	0.0006	0.0032	0.0006	0.0000	0.0000	0.0002	0.0000	0.0002	0.0404	0.0009	0.5820	0.6587
	7	0.0262	0.0002	0.0004	0.0036	0.0007	0.0002	0.0002	0.0002	0.0000	0.0000	0.0001	0.0000	0.0001	0.0001	0.0001	0.0001	0.0001
	8	0.0056	0.0004	0.0004	0.0007	0.0007	0.0004	0.0004	0.0004	0.0000	0.0000	0.0001	0.0000	0.0001	0.0001	0.0001	0.0001	0.0001
516	5	1.249	0.0413	0.0470	0.7034	0.6176	0.0005	0.0068	0.0005	0.0000	0.0000	0.0001	0.0000	0.0001	0.0923	0.0369	0.6567	0.6640
	7	0.0259	0.0002	0.0004	0.0036	0.0005	0.0002	0.0002	0.0002	0.0000	0.0000	0.0001	0.0000	0.0001	0.0001	0.0001	0.0001	0.0001
	8	0.0046	0.0004	0.0004	0.0005	0.0005	0.0004	0.0004	0.0004	0.0000	0.0000	0.0001	0.0000	0.0001	0.0001	0.0001	0.0001	0.0001
516	6	1.249	0.0470	0.0687	0.6991	0.6848	0.0004	0.0079	0.0004	0.0001	0.0001	0.0002	0.0000	0.0002	0.1089	0.0261	0.6592	0.6718
	7	0.0251	0.0002	0.0004	0.0035	0.0004	0.0002	0.0002	0.0002	0.0000	0.0000	0.0001	0.0000	0.0001	0.0001	0.0001	0.0001	0.0001
	8	0.0030	0.0004	0.0004	0.0004	0.0004	0.0004	0.0004	0.0004	0.0000	0.0000	0.0001	0.0000	0.0001	0.0001	0.0001	0.0001	0.0001
516	7	1.249	0.0488	0.0875	0.6902	0.5085	0.0002	0.0088	0.0002	0.0001	0.0001	0.0002	0.0000	0.0002	0.1230	0.0189	0.6698	0.6511
	7	0.0239	0.0002	0.0003	0.0033	0.0002	0.0002	0.0002	0.0002	0.0001	0.0001	0.0001	0.0000	0.0001	0.0001	0.0001	0.0001	0.0001
	8	0.0020	0.0003	0.0003	0.0002	0.0002	0.0003	0.0003	0.0003	0.0000	0.0000	0.0001	0.0000	0.0001	0.0001	0.0001	0.0001	0.0001
516	8	1.248	0.0474	1.4861	0.6963	0.1596	0.0002	0.0090	0.0002	0.0001	0.0001	0.0002	0.0000	0.0002	0.1203	0.0387	0.6885	0.6740
	7	0.0220	0.0002	0.0002	0.0030	0.0000	0.0002	0.0002	0.0002	0.0001	0.0001	0.0001	0.0000	0.0001	0.0001	0.0001	0.0001	0.0001
	8	0.0010	0.0002	0.0002	0.0000	0.0000	0.0002	0.0002	0.0002	0.0001	0.0001	0.0001	0.0000	0.0001	0.0001	0.0001	0.0001	0.0001
517	1	1.249	0.0497	0.0497	0.6881	1.4792	0.0001	0.0022	0.0001	0.0001	0.0001	0.0003	0.0001	0.0003	0.0996	0.3410	0.4836	0.5030
	7	0.0206	0.0002	0.0002	0.0028	0.0001	0.0002	0.0002	0.0002	0.0001	0.0001	0.0001	0.0000	0.0001	0.0001	0.0001	0.0001	0.0001
	8	0.0004	0.0002	0.0002	0.0001	0.0001	0.0002	0.0002	0.0002	0.0001	0.0001	0.0001	0.0000	0.0001	0.0001	0.0001	0.0001	0.0001
517	2	1.249	0.0526	0.0526	0.7044	1.3611	0.0001	0.0023	0.0001	0.0001	0.0001	0.0003	0.0001	0.0003	0.1279	0.3201	0.4143	0.5888
	7	0.0201	0.0002	0.0002	0.0028	0.0001	0.0002	0.0002	0.0002	0.0001	0.0001	0.0001	0.0000	0.0001	0.0001	0.0001	0.0001	0.0001
	8	0.0006	0.0002	0.0002	0.0001	0.0001	0.0002	0.0002	0.0002	0.0001	0.0001	0.0001	0.0000	0.0001	0.0001	0.0001	0.0001	0.0001

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key →																								
517	3	7	1.250	-0.001	0.000	-0.02	33.69	0.0379	0.6971	3.04	-0.07	0.06	-0.03	-0.2468	0.2241												
		8	0.0199	0.0002	0.0002	0.0001	0.0027	-0.0379	1.3961	0.0011	-0.0023	0.0000	0.0000	-0.2800	0.5252												
		9	-0.0007	0.0002	-0.0002	0.0002	-0.0002	-1.6910	0.0023	0.0023	-0.0001	0.0002	0.0002	0.0002	0.0002												
517	4	7	1.250	-0.001	0.000	-0.01	44.99	0.0425	3.03	-0.07	0.06	-0.02	-0.6765	-0.3776													
		8	0.0190	0.0002	0.0002	0.0001	0.0027	-1.4587	1.1222	0.0007	-0.0001	-0.0000	-0.3275	0.5047													
		9	-0.0008	0.0002	-0.0002	0.0002	-0.0001	-1.4587	0.0025	0.0025	-0.0001	0.0002	0.0002	0.0002													
517	5	7	1.249	-0.001	0.000	-0.02	56.19	0.0384	3.04	-0.07	0.06	-0.03	-0.6765	-0.3776													
		8	0.0183	0.0002	0.0002	0.0001	0.0025	-1.3737	0.7069	-0.0001	-0.0001	-0.0001	3.6745	3.7001													
		9	-0.0009	0.0002	-0.0002	0.0002	-0.0001	-1.3737	0.9254	0.0024	-0.0001	0.0002	-0.3763	0.4704													
517	6	7	1.250	-0.001	0.000	-0.01	67.49	0.0412	3.03	-0.08	0.06	-0.02	-1.1072	1.7193													
		8	0.0175	0.0002	0.0002	0.0001	0.0025	-1.3580	0.7167	-0.0007	-0.0001	-0.0002	-0.3675	0.5930													
		9	-0.0010	0.0002	-0.0002	0.0002	-0.0001	-1.3580	0.8377	0.0023	-0.0001	0.0001	0.0001	0.0001													
517	7	7	1.249	-0.001	0.000	-0.01	78.69	0.0439	3.03	-0.08	0.06	-0.03	-0.8213	1.5133													
		8	0.0173	0.0002	0.0002	0.0001	0.0024	-1.8877	0.7185	-0.0010	-0.0001	-0.0003	-0.3891	0.5187													
		9	-0.0007	0.0002	-0.0002	0.0002	-0.0001	-1.8877	1.0490	0.0023	-0.0001	0.0001	0.0001	0.0001													
517	8	7	1.250	-0.001	0.000	-0.00	89.99	0.0373	3.03	-0.08	0.06	-0.03	-0.9123	1.5127													
		8	0.0174	0.0002	0.0002	0.0001	0.0024	-2.9790	0.7063	-0.0011	-0.0002	-0.0003	-0.4366	0.5050													
		9	-0.0004	0.0002	-0.0002	0.0002	-0.0001	-2.9790	1.3371	0.0021	-0.0001	0.0001	0.0001	0.0001													
518	1	7	1.249	-0.001	0.000	-0.03	89.99	0.2237	2.99	-0.08	0.01	-0.02	-2.4910	-2.1187													
		8	0.0082	0.0003	0.0003	0.0002	0.0028	0.0427	0.5172	0.0003	-0.0001	-0.0001	-0.3214	0.4103													
		9	-0.0280	0.0002	-0.0002	0.0002	-0.0037	0.0427	0.6631	0.0020	-0.0001	0.0001	0.0001	0.0001													
518	2	7	1.249	-0.001	0.000	-0.03	78.69	0.2155	2.99	-0.08	0.01	-0.02	-0.3011	0.8777													
		8	0.0078	0.0003	0.0003	0.0002	0.0028	0.0416	0.4943	-0.0046	-0.0002	-0.0003	0.8003	1.1181													
		9	-0.0274	0.0002	-0.0002	0.0002	-0.0036	0.0416	0.6620	-0.0015	-0.0002	-0.0003	0.0003	0.7767													
518	3	7	1.249	-0.001	0.000	-0.03	67.49	0.2317	2.99	-0.09	0.01	-0.03	-0.1984	0.7701													
		8	0.0062	0.0002	0.0002	0.0001	0.0025	0.0482	0.4616	-0.0089	-0.0003	-0.0013	0.2887	0.7767													
		9	-0.0254	0.0002	-0.0002	0.0002	-0.0034	0.0482	0.6802	-0.0058	-0.0003	-0.0009	0.0003	0.7767													
518	4	7	1.250	-0.001	0.000	-0.03	56.19	0.3162	3.00	-0.10	0.02	-0.03	-0.1566	0.7270													
		8	0.0036	0.0002	0.0002	0.0001	0.0022	0.0416	0.3473	-0.0133	-0.0004	-0.0019	0.1936	0.7405													
		9	-0.0224	0.0002	-0.0002	0.0002	-0.0030	0.0416	0.6797	-0.0103	-0.0004	-0.0015	0.0003	0.7405													
518	5	7	1.250	-0.001	0.000	-0.03	44.99	6.1531	3.00	-0.10	0.02	-0.05	-0.1209	0.7079													
		8	0.0001	0.0001	0.0001	0.0001	0.0022	0.0436	8.9225	-0.0175	-0.0004	-0.0024	0.1664	0.7217													
		9	-0.0183	0.0001	-0.0001	0.0001	-0.0025	0.0436	0.6945	-0.0146	-0.0004	-0.0021	0.0004	0.7217													

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																				
518	6	7	1.250	-3.16	33.69	-0.0427	0.9230	3.01	-0.11	0.03	-0.05	0.1071	0.7093	0.0043	-0.0000	0.0007	-0.0212	-0.0004	-0.0030	0.1415	0.6913	
		8	-0.0138	-0.0001	-0.0019	0.0394	0.6953	-0.0186	-0.0005	-0.0005	-0.0025											
518	7	7	1.249	-3.17	22.49	0.0218	0.7548	3.02	-0.11	0.04	-0.05	0.0963	0.6983	0.0092	0.0000	0.0013	-0.0241	-0.0004	-0.0033	0.1389	0.6903	
		8	-0.0091	-0.0000	-0.0013	0.0079	0.7452	-0.0212	-0.0005	-0.0005	-0.0029											
518	8	7	1.250	-3.16	11.19	0.0433	0.7046	3.03	-0.11	0.05	-0.05	0.0960	0.6956	0.0144	0.0001	0.0020	-0.0258	-0.0004	-0.0035	0.1334	0.6900	
		8	-0.0048	0.0001	-0.0007	-0.1161	0.7886	-0.0229	-0.0006	-0.0006	-0.0031											
519	4	7	1.250	-3.20	11.19	0.0528	0.7084	6.06	-0.11	0.05	-0.05	0.0943	0.6962	0.0323	0.0003	0.0045	-0.0259	-0.0004	-0.0036	0.1312	0.7018	
		8	-0.0049	0.0000	-0.0007	-0.0938	0.7481	-0.0231	-0.0006	-0.0006	-0.0032											
519	5	7	1.249	-3.20	22.49	0.0519	0.7133	6.05	-0.11	0.04	-0.05	0.0935	0.6970	0.0267	0.0002	0.0038	-0.0243	-0.0004	-0.0033	0.1408	0.7046	
		8	-0.0096	0.0000	-0.0013	-0.0033	0.6721	-0.0213	-0.0006	-0.0006	-0.0030											
519	6	7	1.250	-3.19	33.69	0.0550	0.7212	6.05	-0.10	0.03	-0.04	0.1051	0.7075	0.0213	0.0002	0.0030	-0.0212	-0.0004	-0.0030	0.1448	0.7109	
		8	-0.0140	-0.0000	-0.0018	0.0337	0.6630	-0.0187	-0.0005	-0.0005	-0.0026											
519	7	7	1.249	-3.19	44.99	0.0471	0.7273	6.04	-0.10	0.02	-0.04	0.1169	0.7045	0.0165	0.0001	0.0024	-0.0175	-0.0004	-0.0024	0.1671	0.7367	
		8	-0.0185	-0.0001	-0.0024	0.0383	0.6729	-0.0146	-0.0004	-0.0004	-0.0021											
519	8	7	1.249	-3.18	56.19	0.0375	0.7397	6.03	-0.09	0.01	-0.03	0.1531	0.7245	0.0127	0.0000	0.0018	-0.0132	-0.0004	-0.0019	0.1952	0.7610	
		8	-0.0227	-0.0001	-0.0029	0.0328	0.6545	-0.0104	-0.0004	-0.0004	-0.0015											
519	9	7	1.249	-3.18	67.49	0.0255	0.7841	6.03	-0.09	0.01	-0.03	0.1952	0.7585	0.0097	0.0000	0.0015	-0.0089	-0.0003	-0.0009	0.2972	0.8304	
		8	-0.0257	-0.0002	-0.0033	0.0391	0.6552	-0.0058	-0.0003	-0.0003	-0.0009											
519	10	7	1.249	-3.17	78.69	0.0107	0.8366	6.03	-0.07	0.00	-0.03	0.3110	0.8751	0.0078	0.0000	0.0013	-0.0045	-0.0002	-0.0007	0.9095	1.4526	
		8	-0.0276	-0.0002	-0.0035	0.0398	0.6485	-0.0014	-0.0002	-0.0002	-0.0004											
519	11	7	1.250	-3.18	89.99	-0.0103	0.8481	6.02	-0.06	0.00	-0.02	-2.1197	-1.7657	0.0072	-0.0000	0.0012	0.0003	-0.0001	-0.0001	-0.2765	0.3285	
		8	-0.0280	-0.0002	-0.0036	-0.0447	0.6560	-0.0020	-0.0001	-0.0001	-0.0001											

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																			
520	1	7	1.250	-0.001	0.0003	11.19	0.0528	6.07	-0.006	0.06	-0.02	0.0004	0.0354	0.6137								
	6	9	0.0361	0.0002	0.0003	0.0050	0.0000	0.7006	0.0033	0.0000	0.0000	0.0000	0.0000	0.3928								
	9		-0.0001			-0.0000	-8.7426	2.1745	0.0017	-0.0001	0.0001	-0.4979	0.3928									
520	2	7	1.250	-0.001	0.0003	22.49	0.0523	6.07	-0.007	0.06	-0.02	0.0003	0.0259	0.5815								
	6	9	0.0359	0.0002	0.0003	0.0050	0.0000	0.7065	0.0032	0.0000	0.0000	0.0000	0.0000	0.5815								
	9		-0.0002			-0.0000	-5.6786	1.3757	0.0017	-0.0001	0.0001	-0.5227	0.3493									
520	3	7	1.249	-0.001	0.0003	33.69	0.0462	6.07	-0.007	0.06	-0.02	0.0002	0.0343	0.4811								
	6	9	0.0356	0.0003	0.0003	0.0050	0.0000	0.7034	0.0027	0.0000	0.0000	0.0000	0.0000	0.4811								
	9		-0.0005			-0.0000	-2.6361	0.8159	0.0018	-0.0001	0.0001	-0.4582	0.2949									
520	4	7	1.249	-0.001	0.0003	44.99	0.0466	6.07	-0.007	0.06	-0.03	0.0001	0.1227	0.4232								
	6	9	0.0346	0.0003	0.0003	0.0049	0.0000	0.7083	0.0021	0.0000	0.0000	0.0000	0.0000	0.4232								
	9		-0.0005			-0.0001	-2.7632	1.2176	0.0018	-0.0001	0.0001	-0.4975	0.3363									
520	5	7	1.250	-0.001	0.0003	56.19	0.0468	6.06	-0.007	0.06	-0.02	0.0000	0.3287	0.2363								
	6	9	0.0340	0.0003	0.0003	0.0048	0.0000	0.7067	0.0017	0.0001	0.0000	0.0000	0.0000	0.2363								
	9		-0.0008			-0.0000	-1.8244	0.4998	0.0018	-0.0002	0.0001	-0.5539	0.2869									
520	6	7	1.250	-0.001	0.0003	67.49	0.0467	6.07	-0.006	0.06	-0.03	0.0000	0.6926	0.1514								
	6	9	0.0334	0.0003	0.0003	0.0047	0.0000	0.7049	0.0009	0.0001	0.0000	0.0000	0.0000	0.1514								
	9		-0.0007			-0.0000	-2.2848	0.6414	0.0016	-0.0002	0.0000	-0.6016	0.1965									
520	7	7	1.252	-0.001	0.0003	78.69	0.0469	6.07	-0.007	0.06	-0.03	0.0000	1.1740	0.7761								
	6	9	0.0332	0.0003	0.0003	0.0046	0.0000	0.7052	0.0005	0.0001	0.0000	0.0000	0.0000	0.7761								
	9		-0.0006			-0.0000	-2.8369	0.4638	0.0015	-0.0001	0.0000	-0.6355	0.0793									
520	8	7	1.250	-0.001	0.0003	89.99	0.0498	6.07	-0.007	0.06	-0.03	0.0000	1.0421	0.4585								
	6	9	0.0330	0.0003	0.0003	0.0046	0.0000	0.7085	0.0008	0.0001	0.0000	0.0000	0.0000	0.4585								
	9		-0.0003			-0.0000	-5.4098	0.6760	0.0013	-0.0002	0.0000	-0.7813	0.0817									
521	1	7	1.249	-0.001	0.0003	11.19	0.0472	6.07	-0.005	0.06	-0.01	0.0015	0.1057	0.6874								
	6	9	0.0378	0.0003	0.0003	0.0052	0.0001	0.6940	0.0113	0.0000	0.0012	0.0000	0.0000	0.6874								
	9		0.0015			0.0001	1.0349	0.3638	0.0087	0.0000	0.0000	0.0402	0.6841									
521	2	7	1.250	-0.001	0.0003	22.49	0.0476	6.08	-0.005	0.06	-0.01	0.0013	0.1022	0.6796								
	6	9	0.0397	0.0003	0.0003	0.0055	0.0003	0.6938	0.0101	0.0000	0.0011	0.0000	0.0000	0.6796								
	9		0.0026			0.0003	0.7157	0.5555	0.0083	0.0000	0.0000	0.0387	0.6764									
521	3	7	1.249	-0.001	0.0003	33.69	0.0458	6.08	-0.005	0.06	-0.01	0.0011	0.0894	0.6572								
	6	9	0.0412	0.0003	0.0003	0.0057	0.0005	0.6976	0.0084	0.0001	0.0011	0.0000	0.0000	0.6572								
	9		0.0040			0.0005	0.5774	0.6429	0.0075	0.0000	0.0000	0.0354	0.6589									

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																				
521	4	7	1.250	1.09	44.99	0.0436	0.6970	-0.068	0.07	0.0000	0.0008	0.0571	0.6500	0.0421	0.0003	0.0058	0.4568	0.6399	0.0063	0.0000	0.0008	0.0269	0.6349
		8	0.0055	0.0005	0.0007																		
521	5	7	1.252	1.10	56.19	0.0426	0.6949	-0.07	0.07	0.0000	-0.01	0.0005	0.6170	0.0427	0.0003	0.0059	0.4056	0.6239	0.0050	0.0000	0.0006	0.0101	0.6199
		8	0.0067	0.0005	0.0008																		
521	6	7	1.250	1.10	67.49	0.0443	0.6960	-0.07	0.08	0.0000	-0.03	-0.0992	0.5052	0.0428	0.0003	0.0059	0.3811	0.6496	0.0039	0.0000	0.0004	-0.1128	0.6290
		8	0.0077	0.0005	0.0010																		
521	7	7	1.249	1.10	78.69	0.0456	0.6973	-0.07	0.08	0.0001	-0.02	-0.3796	0.2201	0.0429	0.0003	0.0059	0.0456	0.6406	0.0027	0.0001	0.0002	-0.2908	0.4066
		8	0.0084	0.0006	0.0010																		
521	8	7	1.251	1.10	89.99	0.0477	0.6973	-0.07	0.08	0.0002	-0.03	-2.3521	0.6970	0.0431	0.0004	0.0060	0.0477	0.6461	0.0008	0.0002	-0.0001	-1.1537	0.2481
		8	0.0087	0.0005	0.0011																		
522	1	7	1.250	3.16	11.20	0.0367	0.607	-0.02	0.06	0.0004	0.00	0.0741	0.6661	0.0409	0.0003	0.0055	0.0367	0.6025	0.0254	0.0002	0.0034	0.0545	0.6712
		8	0.0045	0.0005	0.0005																		
522	2	7	1.249	3.15	22.50	0.0377	0.608	-0.02	0.07	0.0003	0.00	0.0743	0.6653	0.0464	0.0003	0.0063	0.0377	0.6092	0.0261	0.0003	0.0031	0.0636	0.6670
		8	0.0099	0.0006	0.0012																		
522	3	7	1.249	3.16	33.70	0.0422	0.609	-0.03	0.09	0.0002	0.00	0.0681	0.6572	0.0513	0.0004	0.0069	0.0422	0.6512	0.0214	0.0002	0.0028	0.0661	0.6554
		8	0.0148	0.0007	0.0019																		
522	4	7	1.250	3.16	44.99	0.0456	0.610	-0.04	0.10	0.0002	-0.00	0.0669	0.6600	0.0552	0.0005	0.0075	0.0456	0.611	0.0180	0.0002	0.0023	0.0666	0.6481
		8	0.0198	0.0008	0.0026																		
522	5	7	1.248	3.17	56.19	0.0462	0.610	-0.05	0.10	0.0001	-0.01	0.0608	0.6504	0.0582	0.0005	0.0079	0.0462	0.617	0.0135	0.0001	0.0017	0.0539	0.6379
		8	0.0236	0.0008	0.0031																		
522	6	7	1.249	3.17	67.49	0.0450	0.610	-0.06	0.11	0.0000	-0.01	0.0416	0.6476	0.0602	0.0005	0.0081	0.0450	0.6197	0.0087	0.0000	0.0011	0.0263	0.6218
		8	0.0263	0.0008	0.0035																		

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key									
522	1	1.249	5.17	78.69	0.0435	6.11	-0.06	0.11	-0.02	-0.0598	0.6041
	6	0.0611	0.0005	0.0082	0.1560	0.6762	0.0044	-0.0000	0.0005	-0.0800	0.6415
	9	0.0273	0.0008	0.0035	0.0461	0.6566	0.0046	-0.0000	0.0005	-0.0800	0.6415
522	11	1.248	5.15	89.99	0.0461	6.11	-0.06	0.11	-0.03	-45.8563	-45.5719
	6	0.0607	0.0005	0.0083	0.1409	0.6833	0.0000	-0.0001	-0.0001	-1.0187	-0.1039
	9	0.0270	0.0007	0.0036	0.0461	0.6736	0.0011	-0.0002	-0.0000	-1.0187	-0.1039
523	1	1.251	6.28	11.20	0.0358	6.08	0.01	0.09	0.03	0.0652	0.6685
	6	0.0440	0.0003	0.0060	0.3023	0.6895	0.5556	0.0007	0.0074	0.0338	0.6643
	9	0.0111	0.0006	0.0014	0.3023	0.6610	0.0561	0.0003	0.0074	0.0338	0.6643
523	2	1.250	6.27	22.50	0.0410	6.10	0.00	0.10	0.02	0.0653	0.6707
	6	0.0546	0.0004	0.0074	0.1780	0.6795	0.5114	0.0006	0.0069	0.0368	0.6686
	9	0.0228	0.0008	0.0030	0.1780	0.6740	0.0524	0.0003	0.0070	0.0368	0.6686
523	3	1.251	6.29	33.70	0.0392	6.11	-0.00	0.12	0.01	0.0639	0.6671
	6	0.0638	0.0005	0.0085	0.1291	0.6711	0.458	0.0005	0.0061	0.0414	0.6654
	9	0.0336	0.0008	0.0045	0.1291	0.6778	0.0467	0.0003	0.0062	0.0414	0.6654
523	4	1.251	6.29	45.00	0.0397	6.12	-0.00	0.13	0.01	0.0655	0.6636
	6	0.0719	0.0005	0.0095	0.1081	0.6650	0.380	0.0004	0.0050	0.0438	0.6569
	9	0.0427	0.0009	0.0058	0.1081	0.6814	0.0391	0.0003	0.0051	0.0438	0.6569
523	5	1.252	6.28	45.00	0.0395	6.12	-0.01	0.13	0.01	0.0670	0.6667
	6	0.0719	0.0005	0.0095	0.1084	0.6649	0.379	0.0005	0.0050	0.0438	0.6560
	9	0.0427	0.0009	0.0058	0.1084	0.6804	0.0391	0.0003	0.0051	0.0438	0.6560
523	6	1.250	6.29	56.19	0.0403	6.13	-0.02	0.14	0.00	0.0718	0.6577
	6	0.0781	0.0006	0.0102	0.0999	0.6571	0.292	0.0004	0.0038	0.0466	0.6482
	9	0.0493	0.0009	0.0066	0.0999	0.6745	0.0300	0.0002	0.0038	0.0466	0.6482
523	7	1.250	6.30	67.49	0.0421	6.14	-0.04	0.15	-0.00	0.0721	0.6339
	6	0.0829	0.0006	0.0108	0.0938	0.6440	0.196	0.0002	0.0024	0.0471	0.6151
	9	0.0528	0.0009	0.0070	0.0938	0.6712	0.0193	0.0001	0.0023	0.0471	0.6151
523	8	1.249	6.31	76.69	0.0440	6.14	-0.05	0.15	-0.02	0.0676	0.5975
	6	0.0861	0.0007	0.0112	0.0938	0.6537	0.094	0.0001	0.0011	0.0051	0.5750
	9	0.0548	0.0010	0.0073	0.0938	0.6714	0.085	0.0000	0.0009	0.0051	0.5750
523	9	1.251	6.31	89.99	0.0444	6.14	-0.06	0.15	-0.03	-0.9621	-0.8078
	6	0.0874	0.0007	0.0114	0.0921	0.6523	0.006	-0.0001	-0.0001	-3.0978	-2.6838
	9	0.0547	0.0010	0.0073	0.0921	0.6683	-0.0004	-0.0002	-0.0002	-3.0978	-2.6838

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																			
524	1	7	1.250	9.45	11.20	0.0286	6.08	0.04	0.10	0.06	0.0618	0.6514	0.0482	0.0002	0.0066	0.0802	0.0009	0.0104	0.0107	0.0364	0.6421	
		8	0.0167	0.0007	0.0021	0.2098	0.6846	0.0838	0.0006	0.0107												
524	2	7	1.250	9.46	22.51	0.0315	6.11	0.03	0.13	0.06	0.0600	0.6521	0.0634	0.0003	0.0085	0.0749	0.0009	0.0097	0.0103	0.0376	0.6467	
		8	0.0352	0.0008	0.0048	0.1195	0.6893	0.0801	0.0006	0.0103												
524	3	7	1.249	9.45	33.71	0.0342	6.13	0.02	0.15	0.06	0.0606	0.6563	0.0761	0.0005	0.0100	0.0671	0.0008	0.0088	0.0095	0.0397	0.6528	
		8	0.0513	0.0010	0.0069	0.1020	0.6724	0.0734	0.0005	0.0095												
524	4	7	1.250	9.46	45.00	0.0373	6.15	0.00	0.17	0.04	0.0620	0.6630	0.0869	0.0006	0.0112	0.0563	0.0006	0.0074	0.0083	0.0389	0.6572	
		8	0.0636	0.0012	0.0084	0.0957	0.6619	0.0633	0.0004	0.0083												
524	5	7	1.250	9.47	56.19	0.0387	6.16	-0.00	0.18	0.03	0.0622	0.6640	0.0955	0.0007	0.0122	0.0436	0.0005	0.0057	0.0066	0.0350	0.6627	
		8	0.0720	0.0013	0.0094	0.0931	0.6546	0.0499	0.0003	0.0066												
524	6	7	1.252	9.46	67.49	0.0404	6.16	-0.02	0.19	0.01	0.0605	0.6489	0.1019	0.0006	0.0129	0.0291	0.0003	0.0037	0.0042	0.0302	0.6520	
		8	0.0776	0.0013	0.0100	0.0895	0.6494	0.0326	0.0001	0.0042												
524	7	7	1.251	9.47	78.69	0.0418	6.17	-0.05	0.19	-0.01	0.0695	0.6140	0.1061	0.0008	0.0134	0.0138	0.0001	0.0016	0.0018	0.0344	0.6180	
		8	0.0800	0.0014	0.0103	0.0881	0.6492	0.0147	0.0001	0.0018												
524	8	7	1.252	9.46	89.99	0.0444	6.18	-0.06	0.19	-0.03	0.4199	5.7906	0.1083	0.0009	0.0136	-0.0002	-0.0002	-0.0002	-0.0002	0.0302	0.6520	
		8	0.0799	0.0013	0.0103	0.0852	0.6455	-0.0002	0.0002	0.0002												
525	1	7	1.251	12.66	11.20	0.0296	6.09	0.08	0.10	0.09	0.0583	0.6305	0.0521	0.0003	0.0071	0.1000	0.0011	0.0126	0.0131	0.0402	0.6250	
		8	0.0235	0.0007	0.0030	0.1573	0.6559	0.1054	0.0008	0.0131												
525	2	7	1.252	12.65	22.51	0.0313	6.12	0.07	0.14	0.09	0.0571	0.6360	0.0713	0.0004	0.0094	0.0939	0.0010	0.0119	0.0128	0.0416	0.6297	
		8	0.0482	0.0010	0.0064	0.1122	0.6707	0.1022	0.0008	0.0128												
525	3	7	1.250	12.64	33.71	0.0359	6.14	0.06	0.18	0.08	0.0578	0.6418	0.0868	0.0006	0.0112	0.0853	0.0009	0.0109	0.0120	0.0425	0.6350	
		8	0.0672	0.0013	0.0088	0.1007	0.6559	0.0948	0.0008	0.0120												

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key													
525	4	1.251	12.65	45.00	0.0373	6.16	0.03	0.20	0.07	0.0603	0.6496				
	8	0.1001	0.0007	0.0127	0.0915	0.6348	0.0333	0.0008	0.0095	0.0411	0.6428				
	9	0.0821	0.0015	0.0105	0.0915	0.6413	0.0034	0.0006	0.0107						
525	5	1.251	12.65	56.19	0.0384	6.18	0.02	0.21	0.05	0.0641	0.6578				
	8	0.1100	0.0008	0.0137	0.0851	0.6228	0.0583	0.0007	0.0076	0.0398	0.6539				
	9	0.0923	0.0015	0.0116	0.0851	0.6331	0.0674	0.0005	0.0084						
525	6	1.251	12.66	67.48	0.0388	6.19	-0.00	0.23	0.02	0.0663	0.6554				
	8	0.1173	0.0009	0.0144	0.0829	0.6170	0.0401	0.0003	0.0052	0.0355	0.6578				
	9	0.0982	0.0016	0.0122	0.0829	0.6243	0.0470	0.0003	0.0061						
525	7	1.251	12.67	78.69	0.0444	6.20	-0.03	0.23	0.00	0.0824	0.6152				
	8	0.1208	0.0010	0.0147	0.0808	0.6117	0.0190	0.0003	0.0023	0.0196	0.6311				
	9	0.1007	0.0016	0.0125	0.0808	0.6225	0.0217	0.0000	0.0027						
525	8	1.250	12.67	89.99	0.0478	6.20	-0.07	0.23	-0.02	2.1899	3.3837				
	8	0.1233	0.0011	0.0150	0.0782	0.6100	0.0004	-0.0002	-0.0003	2.1899	3.3837				
	9	0.1003	0.0015	0.0124	0.0782	0.6218	0.0000	-0.0002	-0.0001	-14.2568	-10.3456				
526	3	1.248	-3.15	11.19	0.0537	9.08	-0.10	0.05	-0.06	0.0963	0.6945				
	8	0.0485	0.0000	0.0068	0.0902	0.7021	-0.0234	-0.0004	-0.0032	0.1214	0.6854				
	9	-0.0044	0.0005	-0.0006	-0.0902	0.7384	-0.0236	-0.0005	-0.0032						
526	4	1.250	-3.14	22.49	0.0514	9.07	-0.10	0.04	-0.05	0.0992	0.6985				
	8	0.0433	0.0004	0.0060	0.0514	0.7037	-0.0219	-0.0004	-0.0030	0.1290	0.6898				
	9	-0.0088	-0.0000	-0.0012	0.0514	0.6924	-0.0220	-0.0005	-0.0030						
526	5	1.249	-3.13	33.69	0.0542	9.07	-0.09	0.03	-0.05	0.1090	0.6991				
	8	0.0379	0.0004	0.0054	0.0464	0.7116	-0.0190	-0.0004	-0.0026	0.1323	0.6901				
	9	-0.0130	-0.0001	-0.0018	0.0464	0.6909	-0.0194	-0.0005	-0.0026						
526	6	1.248	-3.13	44.99	0.0544	9.08	-0.09	0.03	-0.05	0.1215	0.6942				
	8	0.0330	0.0003	0.0047	0.0457	0.7158	-0.0154	-0.0004	-0.0021	0.1470	0.7063				
	9	-0.0177	-0.0001	-0.0024	0.0457	0.6805	-0.0154	-0.0004	-0.0021						
526	7	1.250	-3.13	56.19	0.0544	9.07	-0.08	0.02	-0.04	0.1628	0.7267				
	8	0.0287	0.0003	0.0041	0.0433	0.7265	-0.0111	-0.0003	-0.0016	0.1688	0.7158				
	9	-0.0216	-0.0001	-0.0028	0.0433	0.6605	-0.0112	-0.0003	-0.0016						
526	8	1.250	-3.12	67.49	0.0516	9.07	-0.08	0.01	-0.03	0.2195	0.7751				
	8	0.0256	0.0002	0.0037	0.0463	0.7300	-0.0070	-0.0003	-0.0010	0.2412	0.7493				
	9	-0.0246	-0.0002	-0.0032	0.0463	0.6616	-0.0066	-0.0003	-0.0010						

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd																				
526	9	7	1.251	-5.13	78.69	0.0486	9.06	-0.07	0.01	-0.03	0.4178	0.9785										
		8	0.0236	0.0002	0.0034	0.0505	0.7389	-0.0025	-0.0002	-0.0005	0.5192	0.9483										
		9	-0.0262	-0.0002	-0.0035		0.6678	-0.0023	-0.0002	-0.0004												
526	10	7	1.251	-5.12	89.99	0.0424	9.06	-0.06	0.01	-0.02	-0.2566	0.3116										
		8	0.0226	0.0001	0.0033	0.0496	0.7414	0.0018	-0.0000	0.0001	-0.4492	0.2684										
		9	-0.0270	-0.0002	-0.0035		0.6611	0.0015	-0.0001	0.0000												
527	1	7	1.250	-0.02	11.19	0.0481	9.11	-0.06	0.06	-0.02	0.0373	0.6105										
		8	0.0541	0.0005	0.0074	0.4503	0.6904	0.0032	0.0000	0.0003	-0.5603	0.4167										
		9	0.0001	0.0002	-0.0000		-1.3545	0.0014	-0.0001	0.0001												
527	2	7	1.251	-0.01	22.49	0.0473	9.11	-0.05	0.06	-0.02	0.0047	0.5736										
		8	0.0539	0.0005	0.0074	-3.7084	0.6928	0.0030	0.0000	0.0003	-0.5516	0.4224										
		9	-0.0003	0.0002	-0.0000		0.6468	0.0013	-0.0001	0.0001												
527	3	7	1.249	-0.01	33.69	0.0468	9.11	-0.05	0.06	-0.02	0.0194	0.5298										
		8	0.0535	0.0005	0.0074	-2.9319	0.6956	0.0025	-0.0000	0.0002	-0.5075	0.3165										
		9	-0.0004	0.0002	-0.0000		0.7166	0.0014	-0.0001	0.0000												
527	4	7	1.250	-0.00	44.99	0.0452	9.10	-0.06	0.06	-0.02	0.1259	0.4632										
		8	0.0528	0.0004	0.0073	-3.0357	0.6947	0.0019	-0.0000	0.0001	-0.5678	0.3232										
		9	-0.0004	0.0002	-0.0000		0.9021	0.0014	-0.0001	0.0000												
527	5	7	1.251	-0.00	56.19	0.0465	9.10	-0.06	0.06	-0.02	0.3406	0.2643										
		8	0.0518	0.0004	0.0072	-2.1195	0.6975	0.0017	-0.0001	0.0000	-0.5969	0.2937										
		9	-0.0006	0.0002	-0.0000		0.7009	0.0014	-0.0001	0.0000												
527	6	7	1.251	-0.00	67.49	0.0456	9.10	-0.06	0.06	-0.02	0.6684	0.1720										
		8	0.0512	0.0004	0.0071	-2.4503	0.7007	0.0010	-0.0001	0.0000	-0.6475	0.1481										
		9	-0.0006	0.0003	-0.0000		0.5779	0.0014	-0.0001	0.0000												
527	7	7	1.251	0.00	78.69	0.0473	9.10	-0.06	0.06	-0.02	0.9560	0.5879										
		8	0.0509	0.0004	0.0071	-3.9968	0.7001	0.0006	-0.0001	0.0000	-0.6914	0.0198										
		9	-0.0003	0.0003	-0.0000		0.8314	0.0013	-0.0001	0.0000												
527	8	7	1.251	0.00	89.99	0.0458	9.10	-0.07	0.06	-0.02	1.3360	0.7412										
		8	0.0508	0.0004	0.0071	-8.9230	0.6992	0.0006	-0.0001	0.0000	-0.8053	0.0889										
		9	-0.0001	0.0003	-0.0000		1.2522	0.0011	-0.0001	0.0000												
528	1	7	1.252	1.02	11.19	0.0448	9.11	-0.04	0.06	-0.01	0.1060	0.6666										
		8	0.0552	0.0004	0.0076	1.3064	0.6879	0.0107	0.0002	0.0014	0.0357	0.6734										
		9	0.0013	0.0003	0.0001		0.5713	0.0082	0.0000	0.0011												

See Key

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																					
528	2	7	1.251	1.03	22.49	0.0446	9.11	-0.04	0.06	-0.01	0.1149	0.6761	0.0027	0.0004	0.0003	0.7549	0.6261	0.0076	0.0000	0.0013	0.0436	0.6736		
		8	0.0572	0.0005	0.0078	0.0000	0.6853	0.0096	0.0002	0.0013	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	
		9	0.0027	0.0004	0.0003	0.0000	0.6261	0.0076	0.0000	0.0010	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	
528	3	7	1.250	1.03	33.69	0.0441	9.11	-0.04	0.07	-0.01	0.0962	0.6487	0.0005	0.0004	0.0005	0.5906	0.6708	0.0069	0.0000	0.0010	0.0423	0.6677	0.0000	
		8	0.0585	0.0005	0.0080	0.0000	0.6857	0.0080	0.0001	0.0001	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
		9	0.0038	0.0004	0.0005	0.0000	0.6708	0.0069	0.0000	0.0009	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
528	4	7	1.250	1.04	44.99	0.0439	9.11	-0.06	0.07	-0.01	0.0617	0.6388	0.0005	0.0005	0.0006	0.4980	0.6837	0.0058	0.0000	0.0008	0.0365	0.6317	0.0000	
		8	0.0593	0.0005	0.0081	0.0000	0.6866	0.0064	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
		9	0.0050	0.0005	0.0006	0.0000	0.6837	0.0058	0.0000	0.0007	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
528	5	7	1.252	1.05	56.19	0.0446	9.12	-0.05	0.07	-0.01	0.0077	0.6098	0.0005	0.0005	0.0008	0.4414	0.6789	0.0047	0.0000	0.0005	-0.0027	0.6188	0.0000	
		8	0.0596	0.0005	0.0081	0.0000	0.6854	0.0048	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
		9	0.0062	0.0005	0.0008	0.0000	0.6789	0.0047	0.0000	0.0005	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
528	6	7	1.251	1.04	67.49	0.0457	9.12	-0.06	0.07	-0.02	0.0659	0.5973	0.0005	0.0005	0.0009	0.4034	0.6918	0.0038	0.0000	0.0003	-0.1262	0.5973	0.0000	
		8	0.0597	0.0005	0.0082	0.0000	0.6864	0.0032	0.0000	0.0003	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
		9	0.0071	0.0005	0.0009	0.0000	0.6918	0.0038	0.0000	0.0004	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
528	7	7	1.254	1.05	78.69	0.0457	9.12	-0.06	0.07	-0.02	0.3832	0.1378	0.0005	0.0005	0.0010	0.3570	0.6777	0.0025	0.0001	0.0000	-0.2900	0.1378	0.0000	
		8	0.0596	0.0005	0.0081	0.0000	0.6860	0.0017	0.0000	0.0001	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
		9	0.0080	0.0005	0.0010	0.0000	0.6777	0.0025	0.0000	0.0001	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
528	8	7	1.253	1.05	89.99	0.0478	9.12	-0.06	0.08	-0.02	0.9751	0.6149	0.0005	0.0005	0.0011	0.3328	0.6660	0.0009	0.0001	0.0000	-1.0558	0.6149	0.0000	
		8	0.0596	0.0005	0.0082	0.0000	0.6880	0.0001	0.0000	0.0001	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
		9	0.0082	0.0005	0.0011	0.0000	0.6660	0.0009	0.0000	0.0001	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
529	1	7	1.253	1.13	11.20	0.0368	9.11	-0.02	0.07	0.00	0.0792	0.6606	0.0004	0.0004	0.0005	0.368	0.6829	0.0286	0.0004	0.0037	0.0511	0.6606	0.0000	
		8	0.0576	0.0004	0.0078	0.0000	0.6829	0.0286	0.0000	0.0004	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
		9	0.0049	0.0004	0.0005	0.0000	0.5931	0.0254	0.0002	0.0033	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
529	2	7	1.251	1.13	22.50	0.0397	9.12	-0.02	0.08	0.00	0.0775	0.6600	0.0005	0.0005	0.0012	0.3144	0.6350	0.0238	0.0002	0.0034	0.0603	0.6600	0.0000	
		8	0.0629	0.0005	0.0085	0.0000	0.6787	0.0258	0.0004	0.0004	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
		9	0.0100	0.0006	0.0012	0.0000	0.6350	0.0238	0.0002	0.0031	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
529	3	7	1.251	1.14	33.70	0.0434	9.13	-0.02	0.09	0.00	0.0715	0.6523	0.0005	0.0005	0.0019	0.2461	0.6508	0.0211	0.0002	0.0028	0.0696	0.6523	0.0000	
		8	0.0673	0.0005	0.0090	0.0000	0.6740	0.0225	0.0003	0.0003	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
		9	0.0149	0.0007	0.0019	0.0000	0.6508	0.0211	0.0002	0.0028	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
529	4	7	1.251	1.15	44.99	0.0450	9.13	-0.03	0.10	0.00	0.0705	0.6482	0.0006	0.0006	0.0026	0.2015	0.6624	0.0177	0.0002	0.0023	0.0642	0.6482	0.0000	
		8	0.0712	0.0006	0.0095	0.0000	0.6712	0.0180	0.0002	0.0023	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
		9	0.0199	0.0008	0.0026	0.0000	0.6624	0.0177	0.0002	0.0023	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key											
529	5	7	1.251	3.15	56.19	0.0447	9.14	-0.03	0.11	-0.00	0.0695	0.6446		
		8	0.0742	0.0006	0.0099	0.1707	0.6672	0.0152	0.0001	0.0017	0.0627	0.6526		
		9	0.0239	0.0008	0.0031	0.1707	0.6671	0.0136	0.0001	0.0017	0.0627	0.6526		
529	6	7	1.251	3.16	67.49	0.0453	9.13	-0.05	0.11	-0.01	0.0557	0.6348		
		8	0.0757	0.0006	0.0101	0.1564	0.6680	0.0084	0.0000	0.0010	0.0415	0.6395		
		9	0.0263	0.0008	0.0035	0.1564	0.6759	0.0088	0.0000	0.0011	0.0415	0.6395		
529	7	7	1.251	3.16	78.69	0.0447	9.14	-0.05	0.11	-0.02	0.0264	0.5561		
		8	0.0767	0.0006	0.0102	0.1520	0.6669	0.0041	0.0000	0.0004	0.0264	0.5561		
		9	0.0273	0.0008	0.0036	0.1520	0.6744	0.0046	0.0000	0.0005	0.0264	0.5561		
529	8	7	1.252	3.17	89.99	0.0447	9.14	-0.07	0.11	-0.02	0.0283	-7.9160		
		8	0.0768	0.0006	0.0102	0.1511	0.6655	0.0001	0.0001	0.0001	0.0283	-7.9160		
		9	0.0267	0.0008	0.0035	0.1511	0.6673	0.0005	0.0002	0.0000	0.0283	-7.9160		
530	1	7	1.251	6.30	11.20	0.0302	9.11	0.01	0.08	0.03	0.0617	0.6640		
		8	0.0609	0.0003	0.0082	0.2833	0.6768	0.0560	0.0006	0.0074	0.0367	0.6711		
		9	0.0116	0.0006	0.0014	0.2833	0.6335	0.0559	0.0004	0.0075	0.0367	0.6711		
530	2	7	1.251	6.30	22.50	0.0360	9.12	0.01	0.10	0.02	0.0614	0.6638		
		8	0.0709	0.0005	0.0094	0.1748	0.6643	0.0520	0.0006	0.0069	0.0387	0.6709		
		9	0.0232	0.0008	0.0030	0.1748	0.6656	0.0525	0.0004	0.0070	0.0387	0.6709		
530	3	7	1.251	6.30	33.70	0.0368	9.14	0.00	0.12	0.02	0.0627	0.6646		
		8	0.0792	0.0005	0.0104	0.1288	0.6565	0.0461	0.0005	0.0061	0.0419	0.6687		
		9	0.0338	0.0008	0.0045	0.1288	0.6775	0.0467	0.0003	0.0062	0.0419	0.6687		
530	4	7	1.250	6.31	45.00	0.0376	9.15	-0.00	0.12	0.02	0.0669	0.6618		
		8	0.0866	0.0006	0.0112	0.1080	0.6498	0.0382	0.0005	0.0050	0.0467	0.6627		
		9	0.0432	0.0009	0.0058	0.1080	0.6768	0.0391	0.0003	0.0051	0.0467	0.6627		
530	5	7	1.251	6.32	56.19	0.0387	9.16	-0.01	0.13	0.01	0.0714	0.6496		
		8	0.0923	0.0007	0.0119	0.0981	0.6445	0.0292	0.0004	0.0038	0.0558	0.6629		
		9	0.0494	0.0009	0.0066	0.0981	0.6673	0.0293	0.0003	0.0038	0.0558	0.6629		
530	6	7	1.250	6.34	67.49	0.0408	9.16	-0.03	0.15	0.00	0.0733	0.6331		
		8	0.0967	0.0007	0.0124	0.0925	0.6409	0.0193	0.0002	0.0024	0.0602	0.6585		
		9	0.0535	0.0009	0.0071	0.0925	0.6691	0.0186	0.0002	0.0024	0.0602	0.6585		
530	7	7	1.251	6.34	78.69	0.0421	9.17	-0.05	0.14	0.01	0.0755	0.5898		
		8	0.0996	0.0008	0.0127	0.0906	0.6396	0.0092	0.0001	0.0010	0.0192	0.6170		
		9	0.0552	0.0010	0.0073	0.0906	0.6683	0.0083	0.0000	0.0010	0.0192	0.6170		

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key													
530	11	7	1.252	6.28	69.99	0.0432	9.16	-0.0003	-0.0003	0.15	-0.0002	-0.0002	-0.0001	0.5757	2.5062	
		8	0.1007	0.0009	0.0128	0.0844	0.6381	-0.0003	-0.0003	-0.0000	-0.0002	-0.0002	-0.0002	3.0822	2.7438	
531	1	7	1.248	9.46	11.20	0.0265	9.11	0.0802	0.05	0.09	0.0010	0.06	0.0634	0.6512		
		8	0.0644	0.0003	0.0087	0.1850	0.6762	0.0839	0.05	0.0006	0.0006	0.0104	0.0390	0.6477		
531	2	7	1.249	9.46	22.51	0.0306	9.13	0.04	0.04	0.12	0.0009	0.06	0.0634	0.6557		
		8	0.0784	0.0004	0.0103	0.1015	0.6611	0.0747	0.0804	0.0006	0.0006	0.0104	0.0390	0.6483		
531	3	7	1.250	9.46	33.71	0.0319	9.15	0.0669	0.03	0.14	0.0008	0.05	0.0649	0.6607		
		8	0.0901	0.0005	0.0116	0.0938	0.6485	0.0737	0.03	0.0005	0.0005	0.0088	0.0395	0.6516		
531	4	7	1.252	9.46	45.00	0.0350	9.16	0.0561	0.02	0.16	0.0007	0.04	0.0672	0.6682		
		8	0.1001	0.0007	0.0127	0.0902	0.6378	0.0633	0.02	0.0005	0.0005	0.0075	0.0395	0.6585		
531	5	7	1.250	9.47	56.19	0.0379	9.17	-0.00	-0.00	0.18	0.0003	0.02	0.0683	0.6692		
		8	0.1077	0.0008	0.0135	0.0874	0.6298	0.0435	0.05	0.0003	0.0003	0.0058	0.0342	0.6586		
531	6	7	1.250	9.47	67.49	0.0394	9.18	-0.02	-0.02	0.19	0.0004	0.00	0.0759	0.6589		
		8	0.1135	0.0008	0.0141	0.0852	0.6237	0.0285	0.0328	0.0002	0.0002	0.0037	0.0334	0.6524		
531	7	7	1.251	9.48	78.69	0.0408	9.19	-0.05	-0.05	0.19	0.0002	-0.00	0.1128	0.6198		
		8	0.1175	0.0009	0.0145	0.0825	0.6186	0.0127	0.0146	0.0001	0.0001	0.0015	0.0395	0.6253		
531	8	7	1.251	9.48	89.99	0.0445	9.20	-0.07	-0.07	0.19	0.0000	-0.02	0.1002	1.2082		
		8	0.1192	0.0010	0.0147	0.0795	0.6181	-0.0016	-0.0006	-0.0002	-0.0002	-0.0004	1.7206	1.1921		
531	9	7	1.251	9.48	89.99	0.0441	9.20	-0.07	-0.07	0.19	0.0000	-0.02	0.1216	1.2299		
		8	0.1192	0.0010	0.0147	0.0798	0.6177	-0.0016	-0.0005	-0.0002	-0.0002	-0.0003	1.7918	2.3384		
532	1	7	1.251	12.65	11.20	0.0254	9.12	0.08	0.10	0.10	0.0011	0.08	0.0598	0.6314		
		8	0.0676	0.0003	0.0091	0.1460	0.6738	0.1000	0.1057	0.0008	0.0008	0.0126	0.0412	0.6265		

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key													
532	2	7	1.253	12.65	22.52	0.0275	0.6343	9.14	0.0936	0.07	0.14	0.011	0.0008	0.0119	0.0596	0.6377
		6	0.0854	0.0004	0.0111	0.1048	0.6343	0.6343	0.0936	0.07	0.14	0.011	0.0008	0.0119	0.0596	0.6377
		9	0.0490	0.0010	0.0065	0.0065	0.6727	0.6727	0.1022	0.1022	0.0008	0.0008	0.0008	0.0129	0.0428	0.6318
532	3	7	1.253	12.64	33.71	0.0307	0.6414	9.17	0.06	0.06	0.17	0.010	0.0008	0.0109	0.0614	0.6451
		6	0.1001	0.0006	0.0128	0.0952	0.6414	0.6414	0.0848	0.0848	0.010	0.010	0.0008	0.0120	0.0445	0.6384
		9	0.0677	0.0012	0.0089	0.0089	0.6581	0.6581	0.0947	0.0947	0.0008	0.0008	0.0008	0.0120	0.0445	0.6384
532	4	7	1.252	12.64	45.00	0.0348	0.6240	9.18	0.04	0.04	0.20	0.009	0.0007	0.006	0.0640	0.6529
		6	0.1115	0.0007	0.0139	0.0870	0.6240	0.6240	0.0727	0.0727	0.009	0.009	0.0007	0.0107	0.0435	0.6487
		9	0.0826	0.0014	0.0106	0.0106	0.6438	0.6438	0.0831	0.0831	0.0007	0.0007	0.0007	0.0107	0.0435	0.6487
532	5	7	1.251	12.66	56.19	0.0380	0.6118	9.20	0.02	0.02	0.21	0.008	0.0005	0.004	0.0716	0.6643
		6	0.1193	0.0009	0.0146	0.0809	0.6118	0.6118	0.0576	0.0576	0.008	0.008	0.0005	0.0076	0.0428	0.6601
		9	0.0926	0.0014	0.0117	0.0117	0.6344	0.6344	0.0673	0.0673	0.0005	0.0005	0.0005	0.0088	0.0428	0.6601
532	6	7	1.250	12.64	67.48	0.0370	0.6084	9.20	-0.00	-0.00	0.22	0.006	0.0003	0.002	0.0827	0.6630
		6	0.1259	0.0009	0.0153	0.0795	0.6084	0.6084	0.0391	0.0391	0.006	0.006	0.0003	0.0051	0.0366	0.6615
		9	0.0985	0.0015	0.0123	0.0123	0.6256	0.6256	0.0470	0.0470	0.0003	0.0003	0.0003	0.0062	0.0366	0.6615
532	7	7	1.250	12.66	78.69	0.0375	0.6059	9.21	-0.03	-0.03	0.22	0.003	0.0001	0.000	0.1086	0.6167
		6	0.1307	0.0009	0.0158	0.0777	0.6059	0.6059	0.0173	0.0173	0.003	0.003	0.0001	0.0027	0.0235	0.6292
		9	0.1012	0.0015	0.0126	0.0126	0.6244	0.6244	0.0215	0.0215	0.0001	0.0001	0.0001	0.0027	0.0235	0.6292
532	8	7	1.251	12.65	89.99	0.0406	0.6045	9.22	-0.07	-0.07	0.22	0.001	-0.0002	-0.0002	0.4772	1.1258
		6	0.1330	0.0010	0.0160	0.0753	0.6045	0.6045	0.0020	0.0020	0.001	0.001	-0.0002	0.0004	0.2197	64.8539
		9	0.1007	0.0015	0.0125	0.0125	0.6231	0.6231	-0.0000	-0.0000	0.0002	0.0002	-0.0002	0.0002	0.2197	64.8539
533	1	7	1.251	-3.15	11.18	0.0526	0.6644	15.15	-0.10	-0.10	0.05	0.004	-0.0004	-0.0035	0.1019	0.7018
		6	0.0817	0.0008	0.0108	0.1326	0.6644	0.6644	0.0235	0.0235	0.004	0.004	-0.0004	0.0035	0.0937	0.6958
		9	-0.0043	0.0001	-0.0006	-0.0006	0.8005	0.8005	-0.0255	-0.0255	0.0004	0.0004	-0.0004	0.0035	0.0937	0.6958
533	2	7	1.253	-3.15	22.49	0.0530	0.6698	15.14	-0.10	-0.10	0.04	0.004	-0.0004	-0.0030	0.1012	0.6986
		6	0.0771	0.0008	0.0103	0.0170	0.6698	0.6698	0.0219	0.0219	0.004	0.004	-0.0004	0.0030	0.1035	0.6968
		9	-0.0089	0.0000	-0.0012	-0.0012	0.6954	0.6954	-0.0234	-0.0234	0.0004	0.0004	-0.0004	0.0032	0.1035	0.6968
533	3	7	1.249	-3.14	33.69	0.0535	0.6768	15.13	-0.09	-0.09	0.02	0.004	-0.0004	-0.0028	0.1124	0.7031
		6	0.0725	0.0007	0.0098	0.0300	0.6768	0.6768	0.0190	0.0190	0.004	0.004	-0.0004	0.0026	0.1124	0.7031
		9	-0.0132	-0.0000	-0.0018	0.0300	0.6793	0.6793	-0.0204	-0.0204	0.0004	0.0004	-0.0004	0.0028	0.1121	0.6987
533	4	7	1.250	-3.12	44.99	0.0521	0.6831	15.12	-0.10	-0.10	0.02	0.004	-0.0004	-0.0023	0.1380	0.7081
		6	0.0681	0.0007	0.0093	0.0326	0.6831	0.6831	0.0151	0.0151	0.004	0.004	-0.0004	0.0021	0.1355	0.7081
		9	-0.0177	-0.0001	-0.0023	0.0326	0.6766	0.6766	-0.0161	-0.0161	0.0004	0.0004	-0.0004	0.0023	0.1355	0.7217

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key											
533	5	7	1.249	-3.11	56.19	0.0493	15.12	-0.0109	0.0003	-0.04	0.1764	0.7327		
	8	8	0.0645	0.0006	0.0088	0.0336	0.6866	-0.0118	-0.0003	-0.0016	0.1552	0.7390		
	9	9	-0.0216	-0.0001	-0.0028	0.0436	0.6619	-0.0118	-0.0003	-0.0017				
533	6	7	1.249	-3.11	67.49	0.0477	15.11	-0.0069	0.0003	-0.02	0.2423	0.7995		
	8	8	0.0614	0.0005	0.0085	0.0454	0.6933	-0.0072	-0.0003	-0.0011	0.2120	0.7854		
	9	9	-0.0242	-0.0002	-0.0032	0.0454	0.6810	-0.0072	-0.0003	-0.0011				
533	7	7	1.252	-3.12	78.69	0.0462	15.11	-0.007	0.01	-0.01	0.5007	1.0227		
	8	8	0.0594	0.0005	0.0082	0.0445	0.6964	-0.0023	-0.0002	-0.0004	0.4074	0.9751		
	9	9	-0.0261	-0.0002	-0.0035	0.0445	0.6694	-0.0026	-0.0002	-0.0005				
533	8	7	1.251	-3.12	89.99	0.0424	15.11	-0.006	0.01	-0.02	-0.2528	0.3512		
	8	8	0.0585	0.0004	0.0081	0.0481	0.6960	0.0021	-0.0001	0.0001	-0.4054	0.1683		
	9	9	-0.0267	-0.0002	-0.0035	0.0481	0.6686	0.0013	-0.0001	0.0000				
534	1	7	1.252	-3.02	11.19	0.0478	15.15	-0.006	0.06	-0.03	0.0410	0.6430		
	8	8	0.0857	0.0008	0.0112	4.9998	0.6577	0.0033	0.0000	0.0004	-0.6123	0.1073		
	9	9	0.0002	0.0002	-0.0000	4.9998	-0.9879	0.0008	-0.0001	0.0000				
534	2	7	1.249	-0.01	22.49	0.0466	15.15	-0.006	0.06	-0.02	0.0204	0.6108		
	8	8	0.0855	0.0007	0.0112	6.4180	0.6576	0.0033	0.0000	0.0004	-0.5255	0.2170		
	9	9	0.0002	0.0002	-0.0000	6.4180	-1.2664	0.0010	-0.0001	0.0000				
534	3	7	1.250	-0.00	33.69	0.0471	15.15	-0.007	0.06	-0.03	-0.0225	0.5447		
	8	8	0.0852	0.0008	0.0112	0.0471	0.6606	0.0028	-0.0000	0.0003	-0.5761	-0.0214		
	9	9	-0.0000	0.0002	-0.0000	-63.6001	16.4008	0.0009	-0.0001	-0.0000				
534	4	7	1.250	0.00	44.99	0.0470	15.15	-0.007	0.06	-0.02	-0.1116	0.4650		
	8	8	0.0847	0.0007	0.0111	0.0470	0.6604	0.0022	-0.0000	0.0002	-0.5978	0.0040		
	9	9	-0.0001	0.0002	-0.0000	-11.5032	2.6304	0.0010	-0.0001	0.0000				
534	5	7	1.252	0.01	56.19	0.0467	15.15	-0.007	0.06	-0.02	-0.3372	0.2438		
	8	8	0.0839	0.0007	0.0111	0.0467	0.6622	0.0018	-0.0001	0.0000	-0.7518	-0.0652		
	9	9	-0.0000	0.0002	-0.0000	-96.6106	26.0820	0.0009	-0.0001	-0.0000				
534	6	7	1.250	0.01	67.49	0.0463	15.15	-0.007	0.06	-0.02	-0.4914	0.0644		
	8	8	0.0836	0.0007	0.0110	0.0463	0.6609	0.0014	-0.0001	0.0000	-0.7927	-0.1583		
	9	9	-0.0002	0.0003	-0.0000	-6.9240	1.3624	0.0008	-0.0001	-0.0000				
534	7	7	1.250	0.01	78.69	0.0474	15.14	-0.006	0.05	-0.02	-0.9270	-0.3353		
	8	8	0.0831	0.0007	0.0110	0.0474	0.6639	0.0007	-0.0001	0.0000	-0.8325	-0.5142		
	9	9	-0.0001	0.0003	-0.0000	-9.4877	0.9652	0.0008	-0.0001	-0.0000				

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd																								
534	0	1.249	0.0001	0.0007	0.0003	0.0110	0.0462	12.5555	0.0003	0.0003	0.0999	0.0462	15.14	0.6337	0.0007	-0.0011	0.0001	0.06	-0.0001	-0.0001	-0.0001	-0.0002	-0.7770	-0.4433	-0.0397
535	1	1.249	0.0018	0.0007	0.0003	0.0114	0.0422	0.9526	0.0003	0.0003	0.1119	0.0422	15.15	0.6567	0.0003	0.0113	0.0002	0.06	-0.0015	0.0011	0.0011	0.0990	0.0600	0.6650	0.6650
535	2	1.249	0.0030	0.0007	0.0004	0.0115	0.0444	0.6656	0.0004	0.0004	0.2249	0.0444	15.15	0.6550	0.0004	0.0100	0.0002	0.06	-0.0013	0.0009	0.0009	0.1004	0.0675	0.6551	0.6551
535	3	1.252	0.0043	0.0008	0.0004	0.0116	0.0459	0.5019	0.0004	0.0004	0.3369	0.0459	15.15	0.6551	0.0008	0.0084	0.0001	0.07	-0.0011	0.0008	0.0008	0.0937	0.0711	0.6498	0.6512
535	4	1.250	0.0055	0.0008	0.0005	0.0117	0.0447	0.4501	0.0005	0.0005	0.4499	0.0447	15.15	0.6531	0.0008	0.0067	0.0000	0.07	-0.0006	0.0006	0.0006	0.0605	0.0734	0.6324	0.6140
535	5	1.252	0.0067	0.0008	0.0005	0.0118	0.0452	0.4098	0.0005	0.0005	0.5619	0.0452	15.15	0.6521	0.0008	0.0050	0.0000	0.07	-0.0005	0.0005	0.0005	0.0036	0.0415	0.5983	0.5999
535	6	1.250	0.0079	0.0008	0.0005	0.0118	0.0452	0.3487	0.0005	0.0005	0.6749	0.0452	15.16	0.6543	0.0008	0.0032	0.0000	0.07	-0.0003	0.0003	0.0003	-0.0853	0.0800	0.4672	0.5835
535	7	1.250	0.0085	0.0008	0.0005	0.0118	0.0467	0.3272	0.0005	0.0005	0.7869	0.0467	15.16	0.6585	0.0008	0.0020	0.0001	0.08	-0.0001	0.0001	0.0001	-0.3542	0.2872	0.1090	0.2459
535	8	1.251	0.0088	0.0008	0.0005	0.0118	0.0464	0.3100	0.0005	0.0005	0.8999	0.0464	15.16	0.6549	0.0008	0.0002	0.0001	0.08	-0.0001	0.0001	0.0001	-23.2845	-2.6979	-2.3198	-2.8291
536	3	1.250	0.0088	0.0006	0.0004	0.0116	0.0355	0.3839	0.0004	0.0004	11.20	0.0355	15.15	0.6547	0.0006	0.0287	0.0002	0.07	0.0004	0.0002	0.0002	0.0860	0.0560	0.6693	0.6611
536	4	1.251	0.0103	0.0007	0.0005	0.0121	0.0396	0.2804	0.0005	0.0005	22.51	0.0396	15.16	0.6545	0.0007	0.0256	0.0003	0.08	0.0004	0.0003	0.0003	0.0871	0.0632	0.6701	0.6620

← See Key →

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd																							
536	5	7	1.250	3.19	33.70	0.0423	15.16	-0.045	0.09	0.00	0.0869	0.6733												
			0.0963	0.0008	0.0125	0.2137	0.494	0.220	0.0003	0.0029	0.0748	0.6631												
			0.0156	0.0006	0.0019		0.6386	0.0214	0.0003	0.0028														
536	6	7	1.250	3.19	45.00	0.0454	15.17	-0.04	0.10	-0.00	0.0895	0.6715												
			0.0993	0.0009	0.0128	0.1932	0.457	0.175	0.0003	0.0023	0.0710	0.6399												
			0.0203	0.0007	0.0027		0.6737	0.0181	0.0002	0.0023														
536	7	7	1.251	3.20	56.19	0.0462	15.17	-0.04	0.10	-0.00	0.0919	0.6632												
			0.1016	0.0009	0.0130	0.1706	0.439	0.127	0.0002	0.0016	0.0707	0.6408												
			0.0239	0.0008	0.0033		0.6895	0.0136	0.0001	0.0017														
536	8	7	1.252	3.21	67.49	0.0443	15.18	-0.05	0.11	-0.01	0.0930	0.6565												
			0.1036	0.0009	0.0132	0.1506	0.385	0.080	0.0001	0.0010	0.0643	0.6167												
			0.0267	0.0008	0.0036		0.6864	0.0087	0.0001	0.0010														
536	9	7	1.252	3.21	78.69	0.0458	15.18	-0.06	0.11	-0.02	0.0933	0.6246												
			0.1042	0.0009	0.0133	0.1288	0.403	0.035	0.0000	0.0004	0.0333	0.5688												
			0.0282	0.0007	0.0037		0.6621	0.0042	-0.0000	0.0004	-0.0333													
536	10	7	1.253	3.21	89.99	0.0453	15.17	-0.06	0.11	-0.03	0.0901	1.5449												
			0.1043	0.0009	0.0133	0.1588	0.405	0.006	-0.0001	-0.0002	0.8901	1.5449												
			0.0268	0.0008	0.0036		0.6874	0.0000	-0.0001	-0.0002	-55.7265	-67.6656												
537	1	7	1.252	6.31	11.20	0.0315	15.15	0.01	0.08	0.02	0.0655	0.6655												
			0.0905	0.0005	0.0118	0.3055	0.556	0.555	0.0007	0.0073	0.0367	0.6648												
			0.0115	0.0007	0.0015		0.6859	0.0561	0.0004	0.0074														
537	2	7	1.252	6.31	22.50	0.0344	15.16	0.00	0.10	0.01	0.0662	0.6684												
			0.0991	0.0006	0.0127	0.1643	0.450	0.514	0.0006	0.0068	0.0385	0.6642												
			0.0235	0.0007	0.0031		0.6667	0.0525	0.0004	0.0069														
537	3	7	1.252	6.31	33.70	0.0387	15.17	0.00	0.12	0.01	0.0681	0.6697												
			0.1055	0.0008	0.0134	0.1212	0.359	0.454	0.0004	0.0062	0.0430	0.6647												
			0.0343	0.0008	0.0046		0.6792	0.0467	0.0004	0.0062														
537	4	7	1.252	6.34	45.00	0.0384	15.18	-0.00	0.13	0.00	0.0742	0.6698												
			0.1116	0.0008	0.0140	0.1023	0.6288	0.377	0.0005	0.0050	0.0451	0.6660												
			0.0437	0.0008	0.0059		0.6761	0.0393	0.0003	0.0051														
537	5	7	1.251	6.33	56.19	0.0401	15.19	-0.02	0.14	0.00	0.0820	0.6605												
			0.1161	0.0009	0.0145	0.1057	0.6252	0.286	0.0004	0.0037	0.0514	0.6536												
			0.0492	0.0010	0.0067		0.6869	0.0296	0.0003	0.0038														

See Key

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																	
537	6	7	1.252	67.49	0.0433	15.19	-0.04	0.15	-0.01	0.1000	0.6507	0.1192	0.0148	0.0433	0.6228	0.0183	0.0003	0.0023	0.0551	0.6507
			0.0535	0.0071	0.0912	0.6711	0.0190	0.0002	0.0023	0.0551	0.6507	0.1192	0.0148	0.0433	0.6228	0.0183	0.0003	0.0023	0.0551	0.6507
			0.0535	0.0071	0.0912	0.6711	0.0190	0.0002	0.0023	0.0551	0.6507	0.1192	0.0148	0.0433	0.6228	0.0183	0.0003	0.0023	0.0551	0.6507
537	7	6	1.253	78.69	0.0455	15.20	-0.05	0.15	-0.02	0.1752	0.5940	0.1214	0.0150	0.0455	0.6186	0.0074	0.0002	0.0006	0.0371	0.5940
			0.0554	0.0073	0.0853	0.6648	0.0080	0.0000	0.0009	0.0371	0.5940	0.1214	0.0150	0.0455	0.6186	0.0074	0.0002	0.0006	0.0371	0.5940
			0.0554	0.0073	0.0853	0.6648	0.0080	0.0000	0.0009	0.0371	0.5940	0.1214	0.0150	0.0455	0.6186	0.0074	0.0002	0.0006	0.0371	0.5940
537	8	7	1.251	89.99	0.0484	15.20	-0.07	0.15	-0.03	0.0482	1.2639	0.1219	0.0150	0.0484	0.6172	0.0018	0.0000	0.0004	0.05204	1.2639
			0.0549	0.0073	0.0899	0.6696	-0.0015	-0.0001	-0.0004	0.05204	1.2639	0.1219	0.0150	0.0484	0.6172	0.0018	0.0000	0.0004	0.05204	1.2639
			0.0549	0.0073	0.0899	0.6696	-0.0015	-0.0001	-0.0004	0.05204	1.2639	0.1219	0.0150	0.0484	0.6172	0.0018	0.0000	0.0004	0.05204	1.2639
538	1	7	1.253	11.20	0.0264	15.15	0.05	0.09	0.05	0.0622	0.6492	0.0927	0.0004	0.0264	0.6570	0.0799	0.0009	0.0103	0.0381	0.6492
			0.0174	0.0023	0.1974	0.6709	0.0836	0.0006	0.0107	0.0381	0.6492	0.0927	0.0004	0.0264	0.6570	0.0799	0.0009	0.0103	0.0381	0.6492
			0.0174	0.0023	0.1974	0.6709	0.0836	0.0006	0.0107	0.0381	0.6492	0.0927	0.0004	0.0264	0.6570	0.0799	0.0009	0.0103	0.0381	0.6492
538	2	7	1.253	22.51	0.0284	15.17	0.03	0.12	0.04	0.0513	0.6511	0.1048	0.0005	0.0284	0.6421	0.0746	0.0009	0.0097	0.0382	0.6511
			0.0361	0.0049	0.1131	0.6890	0.0800	0.0006	0.0103	0.0382	0.6511	0.1048	0.0005	0.0284	0.6421	0.0746	0.0009	0.0097	0.0382	0.6511
			0.0361	0.0049	0.1131	0.6890	0.0800	0.0006	0.0103	0.0382	0.6511	0.1048	0.0005	0.0284	0.6421	0.0746	0.0009	0.0097	0.0382	0.6511
538	3	7	1.252	33.71	0.0320	15.18	0.02	0.15	0.04	0.0632	0.6576	0.1138	0.0007	0.0320	0.6270	0.0667	0.0006	0.0087	0.0413	0.6576
			0.0521	0.0070	0.0982	0.6755	0.0734	0.0006	0.0095	0.0413	0.6576	0.1138	0.0007	0.0320	0.6270	0.0667	0.0006	0.0087	0.0413	0.6576
			0.0521	0.0070	0.0982	0.6755	0.0734	0.0006	0.0095	0.0413	0.6576	0.1138	0.0007	0.0320	0.6270	0.0667	0.0006	0.0087	0.0413	0.6576
538	4	7	1.249	45.01	0.0385	15.20	0.00	0.16	0.03	0.0660	0.6625	0.1202	0.0009	0.0385	0.6162	0.0560	0.0007	0.0074	0.0403	0.6625
			0.0640	0.0012	0.0947	0.6676	0.0633	0.0005	0.0083	0.0403	0.6625	0.1202	0.0009	0.0385	0.6162	0.0560	0.0007	0.0074	0.0403	0.6625
			0.0640	0.0012	0.0947	0.6676	0.0633	0.0005	0.0083	0.0403	0.6625	0.1202	0.0009	0.0385	0.6162	0.0560	0.0007	0.0074	0.0403	0.6625
538	5	7	1.252	56.19	0.0379	15.20	-0.00	0.18	0.02	0.0738	0.6710	0.1261	0.0009	0.0379	0.6099	0.0426	0.0006	0.0057	0.0357	0.6710
			0.0727	0.0094	0.0868	0.6511	0.0496	0.0003	0.0065	0.0357	0.6710	0.1261	0.0009	0.0379	0.6099	0.0426	0.0006	0.0057	0.0357	0.6710
			0.0727	0.0094	0.0868	0.6511	0.0496	0.0003	0.0065	0.0357	0.6710	0.1261	0.0009	0.0379	0.6099	0.0426	0.0006	0.0057	0.0357	0.6710
538	6	7	1.251	67.49	0.0409	15.21	-0.02	0.19	0.00	0.0952	0.6547	0.1295	0.0010	0.0409	0.6042	0.0269	0.0005	0.0042	0.0395	0.6547
			0.0775	0.0101	0.0889	0.6531	0.0320	0.0002	0.0035	0.0395	0.6547	0.1295	0.0010	0.0409	0.6042	0.0269	0.0005	0.0042	0.0395	0.6547
			0.0775	0.0101	0.0889	0.6531	0.0320	0.0002	0.0035	0.0395	0.6547	0.1295	0.0010	0.0409	0.6042	0.0269	0.0005	0.0042	0.0395	0.6547
538	7	6	1.252	78.69	0.0445	15.21	-0.04	0.19	-0.01	0.1603	0.5961	0.1319	0.0159	0.0445	0.6042	0.0104	0.0003	0.0012	0.0466	0.5961
			0.0805	0.0103	0.0841	0.6447	0.0142	0.0001	0.0017	0.0466	0.5961	0.1319	0.0159	0.0445	0.6042	0.0104	0.0003	0.0012	0.0466	0.5961
			0.0805	0.0103	0.0841	0.6447	0.0142	0.0001	0.0017	0.0466	0.5961	0.1319	0.0159	0.0445	0.6042	0.0104	0.0003	0.0012	0.0466	0.5961
538	8	7	1.252	89.99	0.0475	15.22	-0.07	0.19	-0.02	0.1482	0.9851	0.1336	0.0161	0.0475	0.6031	0.0036	0.0001	0.0007	0.1033	0.9851
			0.0802	0.0103	0.0840	0.6458	-0.0023	-0.0000	-0.0005	0.1033	0.9851	0.1336	0.0161	0.0475	0.6031	0.0036	0.0001	0.0007	0.1033	0.9851
			0.0802	0.0103	0.0840	0.6458	-0.0023	-0.0000	-0.0005	0.1033	0.9851	0.1336	0.0161	0.0475	0.6031	0.0036	0.0001	0.0007	0.1033	0.9851

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																					
539	1	7	1.251	12.67	11.21	0.0215	15.15	0.08	0.0998	0.10	0.07	0.0599	0.6315	0.0250	0.0006	0.0032	0.0124	0.0215	0.6536	0.0998	0.0011	0.0126	0.0409	0.6267
539	2	7	1.251	12.68	22.51	0.0255	15.17	0.06	0.0934	0.14	0.08	0.0591	0.6361	0.0496	0.0010	0.0066	0.0138	0.1054	0.6326	0.0934	0.0011	0.0118	0.0427	0.6298
539	3	7	1.252	12.67	33.72	0.0320	15.19	0.06	0.0844	0.17	0.07	0.0616	0.6450	0.196	0.0007	0.0089	0.0147	0.0979	0.6148	0.0844	0.0010	0.0108	0.0450	0.6376
539	4	7	1.251	12.67	45.01	0.0326	15.20	0.03	0.0724	0.19	0.05	0.0658	0.6519	0.1286	0.0008	0.0106	0.0156	0.0878	0.6073	0.0724	0.0009	0.0094	0.0441	0.6469
539	5	7	1.252	12.67	56.19	0.0344	15.21	0.02	0.0566	0.21	0.04	0.0767	0.6653	0.1346	0.0014	0.0117	0.0161	0.0783	0.6283	0.0566	0.0008	0.0075	0.0415	0.6574
539	6	7	1.247	12.66	67.48	0.0398	15.22	0.00	0.0367	0.22	0.01	0.0997	0.6622	0.1389	0.0016	0.0123	0.0166	0.0833	0.6291	0.0367	0.0003	0.0062	0.0355	0.6621
539	7	7	1.252	12.67	78.69	0.0427	15.22	0.04	0.0142	0.22	0.00	0.1232	0.6054	0.1405	0.0012	0.0125	0.0166	0.0755	0.6184	0.0142	0.0001	0.0027	0.0321	0.6429
539	8	7	1.253	12.68	90.00	0.0455	15.23	0.08	0.0042	0.22	0.02	0.4199	0.8468	0.1416	0.0015	0.0125	0.0167	0.0784	0.6258	0.0042	0.0000	0.0005	0.1299	1.1584
540	3	7	0.899	-3.14	11.20	0.2356	-3.03	0.10	-0.0286	0.04	0.05	0.1401	0.6702	-0.0220	-0.0001	-0.0007	-0.0027	0.1072	0.8070	-0.0274	0.0007	-0.0037	0.1414	0.5761
540	4	7	0.897	-3.14	22.50	0.2077	-3.04	0.10	-0.0270	0.04	0.05	0.1404	0.6600	-0.0292	-0.0012	-0.0037	-0.0045	0.0232	0.6374	-0.0270	0.0007	-0.0035	0.1381	0.6803
540	5	7	0.899	-3.14	33.70	0.1911	-3.04	0.09	-0.0238	0.03	0.05	0.1493	0.6719	-0.0174	-0.0001	-0.0021	-0.0045	0.0396	0.6040	-0.0226	0.0006	-0.0030	0.1393	0.6803

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd																						
540	6	7	8	9	0.900	-0.0415	-0.00014	-0.00003	-0.00003	-0.00003	45.00	0.1713	-3.05	0.6289	-0.0200	-0.0005	-0.0005	-0.0027	-0.0027	0.1468	0.6900			
		8	9		-0.0213	-0.00014	-0.00003	-0.00003	-0.00003	-0.00003	-0.00003	0.0791	0.6661	-0.0179	-0.0005	-0.0005	-0.0024	-0.0024	0.1500	0.6960				
540	7	7	8	9	0.899	-0.0450	-0.00015	-0.00004	-0.00004	56.20	0.1756	-3.06	0.6291	-0.0154	-0.0004	-0.0004	-0.0021	-0.0021	0.1588	0.6989				
		8	9		-0.0260	-0.00015	-0.00004	-0.00004	-0.00004	-0.00004	-0.00004	0.0819	0.6557	-0.0125	-0.0004	-0.0004	-0.0018	-0.0018	0.1703	0.7477				
540	8	7	8	9	0.899	-0.0475	-0.00016	-0.00004	-0.00004	67.49	0.1687	-3.06	0.6214	-0.0102	-0.0003	-0.0003	-0.0015	-0.0015	0.1813	0.7359				
		8	9		-0.0295	-0.00016	-0.00004	-0.00004	-0.00004	-0.00004	0.0803	0.6443	-0.0072	-0.0003	-0.0003	-0.0011	-0.0011	0.2400	0.8116					
540	9	7	8	9	0.898	-0.0492	-0.00016	-0.00005	-0.00005	78.69	0.1673	-3.06	0.6228	-0.0049	-0.0002	-0.0002	-0.0008	-0.0008	0.2322	0.8712				
		8	9		-0.0323	-0.00016	-0.00005	-0.00005	-0.00005	-0.00005	0.0832	0.6320	-0.0018	-0.0002	-0.0002	-0.0004	-0.0004	0.7717	1.2260					
540	10	7	8	9	0.898	-0.0505	-0.00015	-0.00005	-0.00005	89.99	0.1558	-3.06	0.6141	-0.0012	-0.0001	-0.0001	-0.0002	-0.0002	-0.3564	0.1138				
		8	9		-0.0332	-0.00015	-0.00005	-0.00005	-0.00005	-0.00005	0.0879	0.6394	0.0024	-0.0001	-0.0001	0.0001	0.0001	-0.3384	0.3229					
541	1	7	8	9	0.898	-0.0149	-0.00006	-0.00002	-0.00002	11.19	0.2260	-3.01	0.6589	-0.0031	-0.0001	-0.0001	-0.0004	-0.0004	0.1629	0.6773				
		8	9		-0.0005	-0.00006	-0.00002	-0.00002	-0.00002	-0.00002	-2.2191	0.1255	0.0029	-0.0001	-0.0001	0.0003	0.0003	-0.2019	0.5610					
541	2	7	8	9	0.898	-0.0160	-0.00006	-0.00002	-0.00002	22.49	0.1996	-3.01	0.6387	-0.0027	-0.0001	-0.0001	-0.0003	-0.0003	0.1572	0.6997				
		8	9		-0.0001	-0.00006	-0.00002	-0.00002	-0.00002	-0.00002	-7.9058	-0.9012	0.0030	-0.0001	-0.0001	0.0003	0.0003	-0.1962	0.5546					
541	3	7	8	9	0.899	-0.0169	-0.00006	-0.00002	-0.00002	33.69	0.1975	-3.02	0.6376	-0.0023	-0.0001	-0.0001	-0.0003	-0.0003	0.1210	0.7254				
		8	9		-0.0005	-0.00006	-0.00002	-0.00002	-0.00002	-0.00002	-2.6713	-0.7063	0.0029	-0.0001	-0.0001	0.0003	0.0003	-0.1926	0.5161					
541	4	7	8	9	0.898	-0.0183	-0.00006	-0.00002	-0.00002	44.99	0.1851	-3.01	0.6245	-0.0017	-0.0001	-0.0001	-0.0002	-0.0002	-0.0455	0.5580				
		8	9		-0.0000	-0.00006	-0.00002	-0.00002	-0.00002	-0.00002	-13.8791	-13.8791	0.0029	-0.0001	-0.0001	0.0002	0.0002	-0.2239	0.4791					
541	5	7	8	9	0.898	-0.0196	-0.00006	-0.00002	-0.00002	56.19	0.1692	-3.02	0.6185	-0.0008	-0.0001	-0.0001	-0.0002	-0.0002	-0.4745	0.0943				
		8	9		-0.0012	-0.00006	-0.00002	-0.00002	-0.00002	-0.00002	-1.2154	-0.3814	0.0027	-0.0001	-0.0001	0.0002	0.0002	-0.2527	0.5122					
541	6	7	8	9	0.898	-0.0194	-0.00007	-0.00002	-0.00002	67.49	0.2013	-3.02	0.6512	-0.0006	-0.0001	-0.0001	-0.0003	-0.0003	14.0428	14.7482				
		8	9		-0.0010	-0.00007	-0.00002	-0.00002	-0.00002	-0.00002	-1.3399	-0.1004	-0.0028	-0.0001	-0.0001	-0.0002	-0.0002	-0.3215	0.4258					

See Key

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key →																				
541	1	8	0.897	-0.007	-0.006	78.89	0.1801	-3.02	-0.006	0.04	-0.002	1.0016	1.7627	0.0006	-0.0002	-0.0001	-0.0001	-0.0002	0.0001	-0.0002	0.0002	-0.4065	0.3588
541	8	9	-0.0196	0.0007	-0.0006	-0.0024	-1.9051	0.6288	0.0026	-0.0002	0.0001	-0.0016	1.7627	-0.0006	0.0002	0.0001	-0.0001	-0.0002	0.0001	-0.0002	0.0002	-0.4065	0.3588
542	A	7	0.899	-0.006	-0.006	89.99	0.2405	-3.02	-0.009	0.05	-0.003	0.7906	1.6586	-0.0009	0.0002	0.0000	-0.0003	-0.0002	0.0000	-0.0003	0.0000	-0.5314	0.2149
542	8	9	-0.0182	0.0006	-0.0006	-0.0025	-1.3518	0.6862	0.0023	-0.0002	0.0001	-0.0016	1.6586	-0.0009	0.0002	0.0000	-0.0003	-0.0002	0.0000	-0.0003	0.0000	-0.5314	0.2149
542	1	7	0.899	-0.005	-0.005	11.19	0.2017	-3.01	-0.04	0.05	-0.001	0.1384	0.7410	0.0011	0.0003	0.0001	-0.0016	0.0001	-0.0016	0.0001	0.0001	0.0532	0.6795
542	8	9	-0.0128	0.0003	-0.0003	-0.0016	1.3696	0.548	0.0127	0.0003	0.0001	-0.0016	0.7410	-0.0011	0.0003	0.0001	-0.0016	0.0001	-0.0016	0.0001	0.0001	0.0532	0.6795
542	2	7	0.900	-0.004	-0.004	22.49	0.2158	-3.01	-0.04	0.05	-0.001	0.1545	0.7495	0.0030	0.0004	0.0001	-0.0015	0.0001	-0.0015	0.0001	0.0001	0.0565	0.6667
542	8	9	-0.0114	0.0004	-0.0004	-0.0015	0.6793	0.6598	0.0111	0.0003	0.0001	-0.0015	0.7495	-0.0030	0.0004	0.0001	-0.0015	0.0001	-0.0015	0.0001	0.0001	0.0565	0.6667
542	3	7	0.895	-0.005	-0.005	33.69	0.2431	-3.01	-0.05	0.06	-0.002	0.1903	0.7476	0.0055	0.0004	0.0001	-0.0013	0.0001	-0.0013	0.0001	0.0001	0.0707	0.6680
542	8	9	-0.0102	0.0004	-0.0004	-0.0009	0.3713	0.6556	0.0087	0.0003	0.0001	-0.0013	0.7476	-0.0055	0.0004	0.0001	-0.0013	0.0001	-0.0013	0.0001	0.0001	0.0707	0.6680
542	4	7	0.898	-0.004	-0.004	44.99	0.2175	-3.01	-0.05	0.07	-0.001	0.2819	0.7197	0.0073	0.0004	0.0001	-0.0011	0.0001	-0.0011	0.0001	0.0001	0.0933	0.6524
542	8	9	-0.0103	0.0004	-0.0004	-0.0012	0.2964	0.6214	0.0058	0.0003	0.0001	-0.0011	0.7197	-0.0073	0.0004	0.0001	-0.0011	0.0001	-0.0011	0.0001	0.0001	0.0933	0.6524
542	5	7	0.898	-0.004	-0.004	56.19	0.2093	-3.01	-0.05	0.07	-0.002	0.2369	0.6131	0.0086	0.0004	0.0001	-0.0008	0.0001	-0.0008	0.0001	0.0001	0.0638	0.6246
542	8	9	-0.0103	0.0004	-0.0004	-0.0013	0.2808	0.5964	0.0040	0.0003	0.0001	-0.0008	0.6131	-0.0086	0.0004	0.0001	-0.0008	0.0001	-0.0008	0.0001	0.0001	0.0638	0.6246
542	6	7	0.899	-0.004	-0.004	67.49	0.2127	-3.01	-0.06	0.06	-0.002	0.0682	0.5119	0.0095	0.0004	0.0001	-0.0006	0.0001	-0.0006	0.0001	0.0001	0.0777	0.5119
542	8	9	-0.0101	0.0004	-0.0004	-0.0014	0.2599	0.6054	0.0024	0.0000	0.0002	-0.0006	0.5119	-0.0095	0.0004	0.0001	-0.0006	0.0001	-0.0006	0.0001	0.0001	0.0777	0.5119
542	7	7	0.899	-0.003	-0.003	78.69	0.1760	-3.01	-0.06	0.06	-0.002	0.0682	0.2545	0.0105	0.0003	0.0001	-0.0003	0.0001	-0.0003	0.0001	0.0001	0.1349	0.4769
542	8	9	-0.0102	0.0004	-0.0004	-0.0014	0.2390	0.5448	0.0040	0.0000	0.0003	-0.0006	0.2545	-0.0105	0.0003	0.0001	-0.0003	0.0001	-0.0003	0.0001	0.0001	0.1349	0.4769
542	8	7	0.899	-0.005	-0.005	89.99	0.2257	-3.01	-0.06	0.06	-0.003	0.0682	1.3525	0.0095	0.0005	0.0002	-0.0003	0.0002	-0.0003	0.0002	0.0002	0.5428	0.1542
542	8	9	-0.0095	0.0005	-0.0005	-0.0015	0.2529	0.5750	0.0013	-0.0002	0.0000	-0.0003	1.3525	-0.0095	0.0005	0.0002	-0.0003	0.0002	-0.0003	0.0002	0.0002	0.5428	0.1542
543	1	7	0.899	-0.006	-0.006	11.19	0.2365	-3.00	-0.01	0.06	-0.001	0.1291	0.6762	0.0089	0.0005	0.0001	-0.0004	0.0001	-0.0004	0.0001	0.0001	0.0842	0.6521
543	8	9	-0.0089	0.0005	-0.0005	-0.0008	0.5430	0.7006	0.0346	0.0008	0.0004	-0.0004	0.6762	-0.0089	0.0005	0.0001	-0.0004	0.0001	-0.0004	0.0001	0.0001	0.0842	0.6521

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

← See Key →

Run	Pt.	Cd																											
543	2	7 9	0.849 -0.0044 -0.0108	5.06 -0.0002 0.0007	22.42 -0.0005 0.0015	0.3277 0.3217	-3.00 0.5697 0.7296	-0.02 0.0307 0.0307	0.07 0.0007 0.0005	0.00 0.0041 0.0040	0.1266 0.0850	0.6776 0.6574																	
543	3	7 9	0.900 -0.0005 -0.0165	3.06 -0.0001 0.0009	33.69 0.0000 0.0024	1.0003 0.2749	-3.00 -0.260 0.7257	-0.03 0.0260 0.0280	0.08 0.0006 0.0004	-0.00 0.0035 0.0036	0.1220 0.0838	0.6756 0.6513																	
543	4	7 9	0.900 0.0028 0.0222	5.05 0.0000 0.0010	44.99 0.0005 0.0031	0.0998 0.2404	-2.99 0.9064 0.7138	-0.03 0.0203 0.0238	0.09 0.0004 0.0004	0.00 0.0027 0.0030	0.1217 0.0868	0.6838 0.6476																	
543	5	7 9	0.900 0.0061 0.0270	3.05 0.0001 0.0011	56.19 0.0008 0.0037	0.1081 0.2123	-2.98 0.7268 0.6966	-0.04 0.0142 0.0188	0.10 0.0003 0.0002	-0.00 0.0019 0.0024	0.1363 0.0777	0.7001 0.6458																	
543	6	7 9	0.899 0.0081 0.0303	5.05 0.0001 0.0012	67.49 0.0011 0.0042	0.0713 0.1978	-2.98 0.6798 0.7032	-0.05 0.0075 0.0132	0.09 0.0003 0.0001	-0.01 0.0010 0.0016	0.2189 0.0541	0.7234 0.6130																	
543	7	7 9	0.901 0.0085 0.0326	5.05 0.0001 0.0011	78.69 0.0012 0.0044	0.1051 0.1781	-2.98 0.7254 0.6744	-0.05 0.0030 0.0069	0.10 0.0000 0.0000	-0.00 0.0003 0.0007	0.1418 0.0163	0.6387 0.5649																	
543	8	7 9	0.901 0.0083 0.0315	5.06 0.0001 0.0011	89.99 0.0012 0.0044	0.0863 0.1860	-2.98 0.7276 0.7046	-0.07 -0.0011 -0.0028	0.10 -0.0002 -0.0003	-0.00 0.0002 0.0000	1.0043 -0.5462	1.2176 0.1339																	
544	1	7 9	0.899 -0.0045 -0.0116	6.19 -0.0003 0.0007	11.19 -0.0005 0.0015	0.3803 0.3327	-3.00 0.6496 0.6764	0.02 0.0640 0.0666	0.07 0.0018 0.0014	0.03 0.0081 0.0083	0.1436 0.1077	0.6369 0.6291																	
544	2	7 9	0.899 0.0036 0.0226	6.18 0.0000 0.0013	22.49 0.0005 0.0032	0.1241 0.3013	-2.99 0.7571 0.7119	0.01 0.0600 0.0622	0.10 0.0016 0.0014	0.03 0.0077 0.0079	0.1351 0.1143	0.6436 0.6405																	
544	3	7 9	0.900 0.0139 0.0358	6.17 0.0003 0.0017	33.69 0.0017 0.0047	0.1262 0.2390	-2.97 0.6190 0.6683	0.00 0.0533 0.0548	0.12 0.0113 0.0013	0.03 0.0069 0.0070	0.1296 0.1257	0.6534 0.6461																	
544	4	7 9	0.899 0.0244 0.0443	6.17 0.0005 0.0021	44.99 0.0030 0.0059	0.1206 0.2377	-2.96 0.6330 0.6707	-0.00 0.0437 0.0451	0.13 0.0011 0.0012	0.02 0.0057 0.0058	0.1320 0.1367	0.6599 0.6448																	

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

← See Key →

Run	Pt.	Cd																					
544	5	7	0.900	6.17	56.19	0.1185	-2.95	-0.02	0.14	0.01	0.1424	0.6488											
		6	0.0334	0.0007	0.0043	0.2291	0.6506	0.0320	0.0009	0.0041	0.1487	0.6400											
		5	0.0508	0.0023	0.0064		0.6366	0.0358	0.0010	0.0043													
544	6	7	0.899	6.17	67.49	0.1052	-2.94	-0.03	0.15	-0.00	0.1605	0.6355											
		6	0.0406	0.0008	0.0052	0.2253	0.6427	0.0204	0.0006	0.0025	0.1377	0.6265											
		5	0.0546	0.0024	0.0071		0.6585	0.0224	0.0006	0.0028													
544	7	7	0.902	6.18	78.69	0.1194	-2.93	-0.05	0.15	-0.01	0.1983	0.6222											
		6	0.0443	0.0010	0.0059	0.1865	0.6662	0.0089	0.0003	0.0011	0.0905	0.5944											
		5	0.0601	0.0022	0.0072		0.5991	0.0105	0.0001	0.0012													
544	8	7	0.900	6.18	89.99	0.1166	-2.93	-0.06	0.15	-0.02	0.9662	9.3756											
		6	0.0453	0.0010	0.0060	0.1938	0.6661	-0.0000	-0.0003	-0.0001	0.9662	-0.3321											
		5	0.0586	0.0022	0.0076		0.6522	-0.0012	-0.0003	-0.0000													
545	1	7	0.899	9.33	11.20	0.5454	-3.00	0.05	0.09	0.06	0.1351	0.6004											
		6	0.0010	-0.0001	0.0000	0.3232	0.3401	0.0882	0.0023	0.0105	0.1158	0.5922											
		5	-0.0159	0.0010	0.0022		0.7068	0.0886	0.0020	0.0105													
545	2	7	0.899	9.31	22.50	0.1085	-2.97	0.04	0.12	0.06	0.1277	0.6081											
		6	0.0130	0.0003	0.0016	0.2827	0.5866	0.0852	0.0021	0.0098	0.1294	0.5915											
		5	0.0331	0.0018	0.0046		0.6962	0.0829	0.0021	0.0098													
545	3	7	0.900	9.30	33.69	0.1312	-2.95	0.04	0.14	0.05	0.1157	0.5293											
		6	0.0296	0.0007	0.0038	0.2559	0.6555	0.0796	0.0018	0.0089	0.1455	0.5946											
		5	0.0481	0.0024	0.0061		0.6377	0.0748	0.0021	0.0089													
545	4	7	0.899	9.30	44.99	0.1025	-2.93	0.02	0.16	0.05	0.1108	0.6429											
		6	0.0478	0.0009	0.0063	0.2321	0.6595	0.0674	0.0014	0.0086	0.1667	0.6022											
		5	0.0601	0.0027	0.0075		0.6240	0.0634	0.0021	0.0076													
545	5	7	0.900	9.30	56.19	0.0846	-2.92	0.00	0.17	0.03	0.1218	0.6546											
		6	0.0631	0.0010	0.0081	0.2154	0.6465	0.0508	0.0012	0.0066	0.1688	0.6252											
		5	0.0689	0.0029	0.0086		0.6259	0.0514	0.0017	0.0064													
545	6	7	0.900	9.30	67.49	0.0952	-2.90	-0.02	0.18	0.01	0.1447	0.6388											
		6	0.0719	0.0013	0.0092	0.1863	0.6434	0.0319	0.0009	0.0040	0.1797	0.6334											
		5	0.0776	0.0028	0.0092		0.5986	0.0338	0.0012	0.0042													
545	7	7	0.901	9.31	78.69	0.1031	-2.89	-0.04	0.18	-0.00	0.1630	0.6008											
		6	0.0760	0.0015	0.0095	0.1711	0.6276	0.0145	0.0004	0.0017	0.1472	0.6112											
		5	0.0822	0.0028	0.0097		0.5951	0.0161	0.0004	0.0019													

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd																									
544	5	7	0.900	6.17	56.19	0.1185	-2.95	-0.02	0.14	0.01	0.1424	0.6488															
		6	0.0334	0.0007	0.0043	0.2291	0.6506	0.0320	0.0009	0.0041	0.1487	0.6400															
		9	0.0508	0.0023	0.0064		0.6366	0.0358	0.0010	0.0043																	
544	6	7	0.899	6.17	67.49	0.1052	-2.94	-0.03	0.15	-0.00	0.1605	0.6355															
		6	0.0406	0.0008	0.0052	0.2253	0.6427	0.0204	0.0006	0.0025	0.1377	0.6265															
		9	0.0546	0.0024	0.0071		0.6585	0.0224	0.0006	0.0028																	
544	7	7	0.902	6.18	78.69	0.1194	-2.93	-0.05	0.15	-0.01	0.1983	0.6222															
		6	0.0443	0.0010	0.0059	0.1865	0.6662	0.0089	0.0003	0.0011	0.0905	0.5944															
		9	0.0601	0.0022	0.0072		0.5991	0.0105	0.0001	0.0012																	
544	8	7	0.900	6.18	89.99	0.1166	-2.93	-0.06	0.15	-0.02	0.9662	9.3756															
		6	0.0453	0.0010	0.0060	0.1938	0.6661	0.0000	0.0001	0.0001	0.1765	-0.3321															
		9	0.0586	0.0022	0.0076		0.6522	0.0012	0.0003	0.0000																	
545	1	7	0.899	9.33	11.20	0.5454	-3.00	0.05	0.09	0.06	0.1351	0.6004															
		6	-0.0010	-0.0001	-0.0000	0.3232	0.3401	0.0882	0.0023	0.0105	0.1158	0.5922															
		9	0.0159	0.0010	0.0022		0.7068	0.0886	0.0020	0.0105																	
545	2	7	0.899	9.31	22.50	0.1085	-2.97	0.04	0.12	0.06	0.1277	0.6081															
		6	0.0130	0.0003	0.0016	0.2827	0.5866	0.0852	0.0021	0.0098	0.1294	0.5915															
		9	0.0331	0.0018	0.0046		0.6962	0.0829	0.0021	0.0098																	
545	3	7	0.900	9.30	33.69	0.1312	-2.95	0.04	0.14	0.05	0.1157	0.6293															
		6	0.0296	0.0007	0.0038	0.2559	0.6555	0.0796	0.0018	0.0100	0.1455	0.5946															
		9	0.0481	0.0024	0.0061		0.6377	0.0748	0.0021	0.0089																	
545	4	7	0.899	9.30	44.99	0.1025	-2.93	0.02	0.16	0.05	0.1108	0.6429															
		6	0.0478	0.0009	0.0063	0.2321	0.6595	0.0674	0.0014	0.0086	0.1667	0.6022															
		9	0.0601	0.0027	0.0075		0.6240	0.0634	0.0021	0.0076																	
545	5	7	0.900	9.30	56.19	0.0846	-2.92	0.00	0.17	0.03	0.1218	0.6546															
		6	0.0631	0.0010	0.0081	0.2154	0.6465	0.0508	0.0012	0.0066	0.1688	0.6252															
		9	0.0689	0.0029	0.0086		0.6259	0.0514	0.0017	0.0064																	
545	6	7	0.900	9.30	67.49	0.0952	-2.90	-0.02	0.18	0.01	0.1447	0.6388															
		6	0.0719	0.0013	0.0092	0.1863	0.6434	0.0319	0.0009	0.0040	0.1797	0.6334															
		9	0.0776	0.0028	0.0092		0.5986	0.0338	0.0012	0.0042																	
545	7	7	0.901	9.31	78.69	0.1031	-2.89	-0.04	0.18	-0.00	0.1630	0.6008															
		6	0.0760	0.0015	0.0095	0.1711	0.6276	0.0145	0.0004	0.0017	0.1472	0.6112															
		9	0.0822	0.0028	0.0097		0.5951	0.0161	0.0004	0.0019																	

See Key

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																			
545	8	0.901	9.32	89.99	0.1093	-2.89	-0.0000	0.19	-0.0001	2.6662	6.8569	0.0770	0.0016	0.0095	0.1093	0.6225	-0.0000	0.19	-0.0001	2.6662	6.8569
	6	0.0847	0.0027	0.0101	0.1609	0.6013	0.0010	-0.0002	-0.0001	-1.1507	-0.6130	0.0847	0.0027	0.0101	0.1609	0.6013	0.0010	-0.0002	-0.0001	-1.1507	-0.6130
547	1	0.901	12.43	11.21	0.0634	-2.97	0.07	0.09	0.06	0.1160	0.5938	0.0025	0.0000	0.0027	0.2973	0.7272	0.1094	0.0025	0.0129	0.0992	0.5821
	6	0.0206	0.0012	0.0027	0.2973	0.6644	0.1087	0.0021	0.0126	0.0992	0.5821	0.0206	0.0012	0.0027	0.2973	0.6644	0.1087	0.0021	0.0126	0.0992	0.5821
547	2	0.900	12.42	22.50	0.1231	-2.95	0.06	0.13	0.07	0.1083	0.5989	0.0232	0.0005	0.0053	0.2761	0.6307	0.1071	0.0023	0.0116	0.1102	0.5748
	6	0.0422	0.0023	0.0053	0.2761	0.6379	0.1012	0.0022	0.0116	0.1102	0.5748	0.0422	0.0023	0.0053	0.2761	0.6379	0.1012	0.0022	0.0116	0.1102	0.5748
547	3	0.900	12.41	33.70	0.1057	-2.92	0.06	0.17	0.07	0.1066	0.6144	0.0464	0.0009	0.0072	0.2413	0.6537	0.1006	0.0023	0.0104	0.1066	0.6144
	6	0.0607	0.0029	0.0072	0.2413	0.5991	0.0911	0.0023	0.0104	0.1066	0.6144	0.0607	0.0029	0.0072	0.2413	0.5991	0.0911	0.0023	0.0104	0.1066	0.6144
547	4	0.900	12.41	44.99	0.0803	-2.90	0.04	0.18	0.06	0.0949	0.6367	0.0708	0.0011	0.0093	0.2019	0.6588	0.0884	0.0016	0.0092	0.0949	0.6367
	6	0.0758	0.0030	0.0088	0.2019	0.5841	0.0789	0.0024	0.0092	0.0949	0.6367	0.0758	0.0030	0.0088	0.2019	0.5841	0.0789	0.0024	0.0092	0.0949	0.6367
547	5	0.901	12.40	56.19	0.0811	-2.87	0.01	0.19	0.04	0.0960	0.6543	0.0881	0.0014	0.0098	0.0811	0.6394	0.0673	0.0012	0.0088	0.0960	0.6543
	6	0.0863	0.0029	0.0098	0.0811	0.5691	0.0637	0.0023	0.0076	0.0960	0.6543	0.0863	0.0029	0.0098	0.0811	0.5691	0.0637	0.0023	0.0076	0.0960	0.6543
547	6	0.901	12.41	67.48	0.0871	-2.86	0.00	0.19	0.03	0.1134	0.6431	0.0984	0.0017	0.0108	0.1440	0.6278	0.0429	0.0017	0.0058	0.1134	0.6431
	6	0.0963	0.0027	0.0108	0.1440	0.5637	0.0464	0.0017	0.0058	0.1134	0.6431	0.0963	0.0027	0.0108	0.1440	0.5637	0.0464	0.0017	0.0058	0.1134	0.6431
547	7	0.900	12.42	78.69	0.0903	-2.85	0.03	0.21	0.00	0.1391	0.6001	0.1030	0.0018	0.0125	0.0903	0.6092	0.0194	0.0005	0.0023	0.1391	0.6001
	6	0.1026	0.0028	0.0125	0.0903	0.5927	0.0225	0.0008	0.0028	0.1391	0.6001	0.1026	0.0028	0.0125	0.0903	0.5927	0.0225	0.0008	0.0028	0.1391	0.6001
547	8	0.900	12.42	90.00	0.1032	-2.85	0.06	0.20	0.02	0.16198	0.6190	0.1014	0.0020	0.0125	0.1032	0.6022	0.0001	0.0000	0.16198	0.6190	
	6	0.1077	0.0027	0.0125	0.1032	0.5822	0.0012	0.0001	0.0000	0.16198	0.6190	0.1077	0.0027	0.0125	0.1032	0.5822	0.0012	0.0001	0.0000	0.16198	0.6190
548	1	0.900	-3.14	11.19	0.5984	-0.03	0.10	0.05	0.06	0.1499	0.6834	-0.0034	-0.0004	-0.0007	0.0191	0.6649	-0.0279	0.0007	0.0038	0.1499	0.6834
	6	-0.0053	-0.0000	-0.0007	0.0191	0.6649	-0.0279	0.0007	0.0038	0.1499	0.6834	-0.0053	-0.0004	-0.0007	0.0191	0.6649	-0.0279	0.0007	0.0038	0.1499	0.6834
548	2	0.901	-3.14	22.49	0.3136	-0.04	0.09	0.03	0.05	0.1618	0.6947	-0.0094	-0.0005	-0.0011	0.0162	0.6753	-0.0254	0.0007	0.0035	0.1618	0.6947
	6	-0.0112	-0.0000	-0.0011	0.0162	0.6753	-0.0263	-0.0007	0.0035	0.1618	0.6947	-0.0112	-0.0000	-0.0011	0.0162	0.6753	-0.0263	-0.0007	0.0035	0.1618	0.6947

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Pt. Cd	See Key																					
548	3	7	0.902	-0.0017	33.69	0.2395	-0.005	-0.0227	0.007	-0.006	0.006	0.1609	0.6931	0.0150	-0.0001	-0.0021	0.0530	0.6358	-0.0232	-0.0006	-0.0031	0.1358	0.6941
548	4	7	0.901	-0.0008	44.99	0.2044	-0.005	-0.0188	0.003	-0.006	-0.005	0.1628	0.7017	-0.0218	-0.0002	-0.0027	0.0559	0.6319	-0.0187	-0.0005	-0.0026	0.1367	0.7057
548	5	7	0.901	-0.0009	56.19	0.1895	-0.006	-0.0140	0.002	-0.004	-0.004	0.1878	0.7430	-0.0250	-0.0003	-0.0031	0.0667	0.6322	-0.0133	-0.0004	-0.0019	0.1510	0.7376
548	6	7	0.900	-0.0010	67.49	0.1769	-0.007	-0.0091	0.001	-0.004	-0.004	0.2264	0.7990	-0.0283	-0.0004	-0.0037	0.0744	0.6378	-0.0079	-0.0003	-0.0012	0.2025	0.8074
548	7	7	0.901	-0.0009	78.69	0.1614	-0.007	-0.0035	0.001	-0.002	-0.002	0.3927	1.0583	-0.0306	-0.0004	-0.0040	0.0764	0.6319	-0.0024	-0.0002	-0.0005	0.5401	1.1258
548	8	7	0.900	-0.0010	89.99	0.1640	-0.007	-0.006	0.000	-0.001	-0.001	0.2841	0.2490	-0.0313	-0.0005	-0.0042	0.0793	0.6295	-0.0019	-0.0001	-0.0000	-0.3944	0.1667
549	1	7	0.898	-0.0003	11.19	-0.0322	-0.002	-0.006	0.005	-0.002	-0.002	0.1531	0.6346	0.0025	0.0000	0.0000	-0.0322	0.3243	-0.0025	-0.0000	-0.0002	0.1500	0.5467
549	2	7	0.899	-0.0003	22.49	-0.1680	-0.002	-0.0035	0.000	-0.000	-0.000	0.1174	0.6373	0.0021	0.0002	-0.0023	0.0000	0.9197	-0.0023	-0.0000	-0.0002	0.1543	0.5910
549	3	7	0.900	-0.0003	33.69	-0.4165	-0.002	-0.005	0.000	-0.000	-0.000	0.0518	0.5900	0.0014	0.0001	0.0000	-0.4165	0.9333	-0.0024	-0.0000	-0.0002	0.1969	0.4850
549	4	7	0.899	-0.0003	44.99	-2.3641	-0.002	-0.006	0.000	-0.000	-0.000	0.0272	0.5512	0.0002	0.0001	0.0000	-2.3641	3.9432	-0.0024	-0.0001	-0.0002	0.2197	0.4639
549	5	7	0.899	-0.0003	56.19	-1.5695	-0.002	-0.006	0.000	-0.000	-0.000	0.2501	0.4128	0.0005	0.0001	0.0000	-1.5695	-5.5577	-0.0024	-0.0001	-0.0001	0.2730	0.4128

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key														
549	6	7	0.500	-0.0007	-0.0002	67.49	1.4764	-0.3326	-0.0007	0.0022	0.04	-0.0000	-0.0000	-0.0000	-0.9110	-0.3713
549	7	6	0.900	-0.0008	-0.0002	78.69	1.1799	-0.4560	0.0000	0.05	-0.0001	-0.0001	-0.0001	-43.6106	-39.4044	
549	8	6	0.901	-0.0004	-0.0003	89.99	2.6843	-0.9330	-0.0001	0.05	-0.0002	-0.0000	-0.0000	6.2085	7.5355	
550	1	6	0.900	0.0048	0.0003	11.19	0.0845	0.6913	-0.0001	0.06	0.0003	0.0019	0.0014	0.1212	0.7190	
550	2	6	0.899	0.0063	0.0003	22.49	0.0741	0.6703	0.0102	0.05	0.0003	0.0017	0.0014	0.1348	0.7224	
550	3	6	0.899	0.0065	0.0003	33.69	0.0898	0.7124	0.0092	0.06	0.0003	0.0014	0.0012	0.1669	0.7324	
550	4	6	0.900	0.0074	0.0004	44.99	0.1095	0.7136	0.0066	0.07	0.0002	0.0009	0.0010	0.2240	0.6980	
550	5	6	0.899	0.0074	0.0004	56.19	0.1208	0.7418	0.0046	0.06	0.0001	0.0005	0.0007	0.1946	0.6275	
550	6	6	0.899	0.0076	0.0005	67.49	0.0934	0.7136	0.0027	0.07	0.0000	0.0003	0.0005	0.0360	0.5578	
550	7	6	0.898	0.0070	0.0004	78.69	0.1965	0.8564	0.0014	0.07	-0.0001	0.0000	0.0003	-0.4184	0.1670	
550	8	6	0.900	0.0077	0.0005	89.99	0.1003	0.7390	-0.0005	0.07	-0.0002	-0.0000	-0.0000	1.7990	2.3456	
			0.0104	0.0001	0.0005	0.0014	0.2427	0.7119	-0.0020	-0.0002	-0.0002	-0.0000	-0.0000	-0.6756	-0.0367	

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd																			
551	1	7	0.899	3.07	11.19	0.1287	-0.00	-0.01	0.06	0.00	0.1256	0.6762								
		6	0.0077	0.0001	0.0010	0.4048	0.6867	0.0350	0.0008	0.0047	0.0934	0.6608								
		9	0.0057	0.0004	0.0007	0.0000	0.6956	0.0320	0.0005	0.0042										
551	2	7	0.898	3.07	22.49	0.1220	-0.00	-0.02	0.07	0.00	0.1227	0.6727								
		6	0.0133	0.0003	0.0018	0.2922	0.6790	0.0313	0.0007	0.0042	0.0920	0.6490								
		9	0.0117	0.0006	0.0015	0.0000	0.6494	0.0306	0.0005	0.0039										
551	3	7	0.899	3.07	33.69	0.0698	0.00	-0.03	0.08	0.00	0.1230	0.6765								
		6	0.0195	0.0002	0.0024	0.2487	0.6370	0.0262	0.0006	0.0035	0.0954	0.6462								
		9	0.0178	0.0008	0.0023	0.0000	0.6621	0.0269	0.0005	0.0034										
551	4	7	0.899	3.06	44.99	0.0811	0.01	-0.03	0.09	0.00	0.1201	0.6694								
		6	0.0244	0.0003	0.0032	0.2225	0.6560	0.0209	0.0005	0.0028	0.1043	0.6597								
		9	0.0233	0.0010	0.0031	0.0000	0.6704	0.0227	0.0004	0.0029										
551	5	7	0.899	3.07	56.19	0.0884	0.01	-0.04	0.10	0.00	0.1325	0.6853								
		6	0.0279	0.0004	0.0037	0.2015	0.6715	0.0145	0.0003	0.0019	0.0926	0.6461								
		9	0.0280	0.0011	0.0037	0.0000	0.6684	0.0178	0.0003	0.0023										
551	6	7	0.899	3.08	67.49	0.0986	0.02	-0.04	0.11	0.00	0.1802	0.6998								
		6	0.0298	0.0005	0.0040	0.1890	0.6821	0.0083	0.0003	0.0011	0.0895	0.6558								
		9	0.0312	0.0011	0.0042	0.0000	0.6784	0.0118	0.0002	0.0015										
551	7	7	0.900	3.07	78.69	0.0913	0.02	-0.05	0.10	0.00	0.1060	0.6517								
		6	0.0308	0.0005	0.0042	0.1706	0.6819	0.0034	0.0000	0.0004	0.0275	0.5597								
		9	0.0336	0.0011	0.0043	0.0000	0.6505	0.0062	0.0000	0.0006										
551	8	7	0.900	3.07	89.99	0.0825	0.02	-0.06	0.10	0.00	2.7808	2.3120								
		6	0.0306	0.0005	0.0041	0.1640	0.6780	-0.0004	-0.0002	-0.0002	-0.8096	0.0018								
		9	0.0334	0.0010	0.0042	0.0000	0.6415	0.0018	-0.0003	0.0000										
552	1	7	0.899	6.19	11.19	0.0916	-0.00	0.01	0.08	0.03	0.1326	0.6392								
		6	0.0124	0.0002	0.0015	0.3602	0.6381	0.0649	0.0017	0.0083	0.1136	0.6311								
		9	0.0115	0.0008	0.0016	0.0000	0.7159	0.0654	0.0014	0.0082										
552	2	7	0.900	6.19	22.49	0.0803	0.01	0.00	0.10	0.04	0.1276	0.6451								
		6	0.0244	0.0003	0.0029	0.2867	0.6137	0.0606	0.0015	0.0078	0.1248	0.6464								
		9	0.0235	0.0013	0.0031	0.0000	0.6761	0.0605	0.0015	0.0078										
552	3	7	0.899	6.18	33.70	0.1021	0.02	0.00	0.12	0.03	0.1229	0.6592								
		6	0.0353	0.0007	0.0046	0.2443	0.6526	0.0539	0.0013	0.0071	0.1371	0.6492								
		9	0.0358	0.0017	0.0048	0.0000	0.6710	0.0533	0.0014	0.0069										

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key											
552	4	0.899	6.18	44.99	0.0933	0.03	-0.00	0.13	0.01	0.1275	0.6574	0.1518	0.6544
	6	0.0474	0.0008	0.0062	0.2326	0.6579	0.0437	0.0011	0.0057	0.0056	0.6574	0.1518	0.6544
	9	0.0451	0.0021	0.0058	0.2326	0.6501	0.0434	0.0013	0.0056	0.0056	0.6574	0.1518	0.6544
552	5	0.899	6.17	56.19	0.0948	0.05	-0.01	0.14	0.01	0.1383	0.6446	0.1650	0.6489
	6	0.0566	0.0010	0.0074	0.2343	0.6581	0.0320	0.0008	0.0041	0.0041	0.6446	0.1650	0.6489
	9	0.0506	0.0023	0.0064	0.2343	0.6410	0.0321	0.0010	0.0041	0.0041	0.6446	0.1650	0.6489
552	6	0.899	6.19	67.49	0.0981	0.06	-0.03	0.15	-0.00	0.1530	0.6346	0.1484	0.6357
	6	0.0627	0.0012	0.0081	0.2130	0.6482	0.0206	0.0006	0.0026	0.0026	0.6346	0.1484	0.6357
	9	0.0563	0.0023	0.0070	0.2130	0.6228	0.0212	0.0006	0.0026	0.0026	0.6346	0.1484	0.6357
552	7	0.899	6.19	78.69	0.1074	0.06	-0.04	0.15	-0.01	0.1642	0.5998	0.0997	0.5826
	6	0.0655	0.0014	0.0084	0.1934	0.6423	0.0095	0.0003	0.0011	0.0011	0.5998	0.0997	0.5826
	9	0.0596	0.0023	0.0072	0.1934	0.6090	0.0095	0.0001	0.0011	0.0011	0.5998	0.0997	0.5826
552	8	0.900	6.19	89.99	0.1133	0.06	-0.06	0.15	-0.03	-3.9386	-3.0681	-4.2850	-2.9095
	6	0.0654	0.0014	0.0084	0.1133	0.6440	0.0001	-0.0001	-0.0002	-3.9386	-3.0681	-4.2850	-2.9095
	9	0.0608	0.0021	0.0074	0.1133	0.6121	0.0003	-0.0002	-0.0002	-3.9386	-3.0681	-4.2850	-2.9095
553	1	0.899	9.33	11.20	0.0847	0.00	0.05	0.09	0.07	0.1296	0.6045	0.1211	0.5911
	6	0.0171	0.0002	0.0021	0.3244	0.6382	0.0894	0.0023	0.0108	0.1296	0.6045	0.1211	0.5911
	9	0.0162	0.0010	0.0022	0.3244	0.6846	0.0872	0.0021	0.0103	0.1296	0.6045	0.1211	0.5911
553	2	0.898	9.32	22.50	0.0957	0.02	0.04	0.12	0.06	0.1208	0.6150	0.1370	0.5928
	6	0.0346	0.0006	0.0044	0.0957	0.6468	0.0670	0.0021	0.0107	0.1208	0.6150	0.1370	0.5928
	9	0.0348	0.0018	0.0044	0.2602	0.6424	0.0813	0.0022	0.0096	0.1208	0.6150	0.1370	0.5928
553	3	0.899	9.32	33.70	0.0865	0.04	0.03	0.14	0.06	0.1085	0.6322	0.1539	0.5956
	6	0.0535	0.0009	0.0070	0.0865	0.6625	0.0807	0.0017	0.0102	0.1085	0.6322	0.1539	0.5956
	9	0.0483	0.0025	0.0061	0.2609	0.6345	0.0731	0.0022	0.0087	0.1085	0.6322	0.1539	0.5956
553	4	0.899	9.32	44.99	0.0724	0.06	0.02	0.16	0.05	0.1041	0.6460	0.1747	0.6053
	6	0.0712	0.0010	0.0091	0.0724	0.6414	0.0681	0.0014	0.0068	0.1041	0.6460	0.1747	0.6053
	9	0.0603	0.0028	0.0074	0.2336	0.6179	0.0618	0.0021	0.0074	0.1041	0.6460	0.1747	0.6053
553	5	0.900	9.32	56.19	0.0858	0.08	-0.00	0.17	0.03	0.1156	0.6518	0.1768	0.6297
	6	0.0822	0.0014	0.0104	0.0858	0.6343	0.0505	0.0011	0.0065	0.1156	0.6518	0.1768	0.6297
	9	0.0704	0.0029	0.0084	0.2078	0.6026	0.0498	0.0017	0.0062	0.1156	0.6518	0.1768	0.6297
553	6	0.900	9.31	67.49	0.0947	0.09	-0.01	0.18	0.02	0.1346	0.6333	0.1874	0.6344
	6	0.0886	0.0016	0.0109	0.0947	0.6197	0.0318	0.0008	0.0040	0.1346	0.6333	0.1874	0.6344
	9	0.0781	0.0028	0.0091	0.1833	0.5863	0.0322	0.0012	0.0040	0.1346	0.6333	0.1874	0.6344

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key									
553	1	0.900	9.52	78.69	0.1043	0.09	-0.04	0.19	-0.01	0.1382	0.5856
	6	0.0909	0.0019	0.0110	0.6073	0.0148	0.0004	0.0004	0.0017	0.1544	0.6053
	9	0.0824	0.0028	0.0098	0.5962	0.0150	0.0004	0.0004	0.0018		
553	8	0.900	9.34	89.99	0.1180	0.10	-0.06	0.19	-0.03		
	6	0.0895	0.0021	0.0107	0.6007	0.0000	-0.0001	0.0001	-0.0001	-19.5952	-26.9238
	9	0.0858	0.0026	0.0100	0.5880	0.0000	-0.0002	-0.0002	-0.0002	-40.5880	-39.1651
554	1	0.900	12.43	11.20	0.0812	0.00	0.07	0.09	0.07	0.1120	0.5939
	6	0.0217	0.0003	0.0026	0.6191	0.1107	0.0024	0.0024	0.0131	0.1014	0.5772
	9	0.0202	0.0013	0.0028	0.6979	0.1072	0.0021	0.0021	0.0123		
554	2	0.899	12.42	22.50	0.0818	0.03	0.06	0.14	0.07	0.1052	0.5991
	6	0.0453	0.0007	0.0058	0.6469	0.1081	0.0022	0.0022	0.0129	0.1145	0.5703
	9	0.0427	0.0023	0.0053	0.6294	0.0994	0.0023	0.0023	0.0113		
554	3	0.898	12.42	33.70	0.0682	0.06	0.05	0.16	0.06	0.1017	0.6188
	6	0.0702	0.0009	0.0092	0.6614	0.1022	0.0020	0.0020	0.0126	0.1319	0.5715
	9	0.0604	0.0030	0.0073	0.6082	0.0900	0.0023	0.0023	0.0102		
554	4	0.900	12.42	44.99	0.0722	0.08	0.04	0.18	0.06	0.0908	0.6402
	6	0.0911	0.0013	0.0117	0.6421	0.0880	0.0016	0.0016	0.0112	0.0898	0.6473
	9	0.0761	0.0030	0.0088	0.5811	0.0773	0.0024	0.0024	0.0089	0.1874	0.5947
554	5	0.900	12.42	56.19	0.0858	0.10	0.01	0.19	0.05	0.0908	0.6402
	6	0.1033	0.0017	0.0128	0.6201	0.0673	0.0012	0.0012	0.0087	0.0898	0.6473
	9	0.0869	0.0029	0.0098	0.5642	0.0627	0.0023	0.0023	0.0074	0.1596	0.5809
554	6	0.899	12.42	67.48	0.0917	0.11	-0.00	0.20	0.03	0.0908	0.6473
	6	0.1091	0.0020	0.0131	0.6043	0.0424	0.0009	0.0009	0.0054	0.1064	0.6378
	9	0.0955	0.0028	0.0110	0.5782	0.0448	0.0017	0.0017	0.0056	0.1959	0.6344
554	7	0.899	12.43	78.69	0.0949	0.12	-0.04	0.20	0.00	0.1331	0.5913
	6	0.1115	0.0021	0.0131	0.5890	0.0188	0.0007	0.0007	0.0022	0.1881	0.6327
	9	0.1029	0.0028	0.0120	0.5858	0.0212	0.0007	0.0007	0.0026		
554	8	0.901	12.43	90.00	0.1028	0.12	-0.06	0.20	-0.03	3.6269	4.2986
	6	0.1098	0.0022	0.0129	0.5894	0.0002	-0.0001	0.0001	-0.0002	-3.6977	-2.7970
	9	0.1080	0.0027	0.0125	0.5794	0.0003	-0.0002	-0.0002	-0.0002		
555	1	0.901	-3.14	11.19	0.3281	1.00	-0.10	0.05	-0.06	0.1506	0.6881
	6	0.0024	-0.0001	0.0004	0.6704	-0.0270	-0.0008	-0.0008	-0.0037	0.1400	0.6954
	9	-0.0051	0.0000	-0.0007	0.6972	-0.0279	-0.0007	-0.0007	-0.0038		

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key											
555	2	7	0.901	-3.14	22.49	0.7674	0.99	-0.09	0.04	-0.05	0.1607	0.6900	
		8	-0.0028	-0.0004	-0.0002	0.0114	0.4635	-0.0251	-0.0008	-0.0034	0.1348	0.6924	
		9	-0.0116	0.0000	-0.0014	-0.0114	0.6287	-0.0265	-0.0007	-0.0036			
555	3	7	0.901	-3.14	33.69	0.3258	0.99	-0.09	0.03	-0.05	0.1665	0.6982	
		8	-0.0083	-0.0005	-0.0009	0.0385	0.5720	-0.0224	-0.0007	-0.0031	0.1356	0.6976	
		9	-0.0164	-0.0001	-0.0021	0.0385	0.6540	-0.0234	-0.0006	-0.0032			
555	4	7	0.900	-3.13	44.99	0.2517	0.98	-0.09	0.02	-0.05	0.1710	0.7141	
		8	-0.0132	-0.0006	-0.0017	0.0627	0.6423	-0.0184	-0.0006	-0.0026	0.1356	0.7029	
		9	-0.0210	-0.0002	-0.0028	0.0627	0.6696	-0.0191	-0.0005	-0.0026			
555	5	7	0.899	-3.14	56.19	0.1995	0.97	-0.08	0.02	-0.04	0.1937	0.7459	
		8	-0.0179	-0.0007	-0.0022	0.0665	0.6280	-0.0138	-0.0004	-0.0020	0.1514	0.7391	
		9	-0.0261	-0.0003	-0.0033	0.0665	0.6432	-0.0136	-0.0004	-0.0020			
555	6	7	0.900	-3.13	67.49	0.2075	0.97	-0.07	0.01	-0.03	0.2357	0.8074	
		8	-0.0205	-0.0008	-0.0026	0.0641	0.6457	-0.0087	-0.0004	-0.0014	0.2015	0.8130	
		9	-0.0300	-0.0003	-0.0037	0.0641	0.6235	-0.0081	-0.0003	-0.0013			
555	7	7	0.901	-3.13	78.69	0.1767	0.97	-0.07	0.01	-0.03	0.4305	1.0880	
		8	-0.0231	-0.0008	-0.0028	0.0842	0.6152	-0.0033	-0.0002	-0.0007	0.5363	1.1825	
		9	-0.0313	-0.0005	-0.0042	0.0842	0.6706	-0.0026	-0.0002	-0.0006			
555	8	7	0.901	-3.14	89.99	0.1756	0.97	-0.06	0.01	-0.03	0.2964	0.2677	
		8	-0.0240	-0.0008	-0.0029	0.0794	0.6227	0.0022	-0.0001	0.0001	-0.4518	0.0445	
		9	-0.0332	-0.0005	-0.0042	0.0794	0.6449	0.0018	-0.0001	0.0000			
556	1	7	0.901	-0.03	11.19	0.1279	1.02	-0.05	0.05	-0.02	0.1696	0.6652	
		8	0.0089	0.0002	0.0012	-4.0500	0.7027	0.0039	0.0001	0.0005	0.2000	0.5538	
		9	-0.0003	0.0002	-0.0000	-4.0500	0.1546	0.0022	-0.0000	0.0002			
556	2	7	0.898	-0.02	22.49	0.0759	1.01	-0.06	0.05	-0.02	0.1290	0.6434	
		8	0.0086	0.0001	0.0011	-4.0953	0.6518	0.0037	0.0000	0.0004	-0.2041	0.4887	
		9	-0.0003	0.0003	0.0000	-4.0953	-0.9964	0.0023	-0.0000	0.0002			
556	3	7	0.899	-0.02	33.69	0.0846	1.01	-0.06	0.06	-0.02	0.0567	0.5726	
		8	0.0075	0.0001	0.0010	3.0099	0.6925	0.0033	0.0000	0.0003	0.2318	0.4622	
		9	0.0004	0.0002	-0.0000	3.0099	-0.2704	0.0023	-0.0001	0.0002			
556	4	7	0.899	-0.02	44.99	0.1513	1.01	-0.05	0.05	-0.03	0.0471	0.5668	
		8	0.0059	0.0001	0.0009	0.1513	0.8427	0.0023	-0.0000	0.0002	-0.2849	0.5150	
		9	-0.0000	0.0002	0.0000	-26.4143	-1.6322	0.0022	-0.0001	0.0001			

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key											
556	5	7	0.899	-0.03	56.19	0.0699	1.01	-0.06	0.05	-0.02	0.0001	-0.2621	0.3673
		6	0.0054	0.0000	0.0008	0.0699	0.7659	0.0016	0.0000	0.0001	0.0001	-0.3419	0.2986
		9	0.0000	0.0002	-0.0000	21.0760	-1.6224	0.0021	-0.0001	0.0000	0.0000		
556	6	7	0.900	-0.03	67.49	0.1086	1.01	-0.06	0.05	-0.02	0.0000	-1.0819	-0.2793
		8	0.0045	0.0000	0.0007	0.1086	0.8558	0.0007	0.0001	-0.0000	0.0000	-0.4198	0.2544
		9	-0.0007	0.0003	0.0000	-2.3106	-0.2981	0.0020	-0.0001	0.0000	0.0000		
556	7	7	0.899	-0.03	78.69	-0.0041	1.01	-0.06	0.05	-0.02	0.0000	-3.0985	-2.1680
		8	0.0054	0.0000	0.0007	-3.9872	0.6522	0.0003	0.0001	-0.0000	0.0000	-0.5657	0.0822
		9	-0.0003	0.0003	0.0000	-3.9872	-0.3383	0.0019	-0.0002	0.0000	0.0000		
556	8	7	0.900	-0.03	89.99	0.0544	1.01	-0.06	0.05	-0.02	0.0000	-8.5816	-8.6291
		8	0.0053	0.0000	0.0008	0.0544	0.7569	0.0001	0.0001	-0.0000	-0.0000	-0.8045	-0.1422
		9	-0.0008	0.0003	0.0000	-1.8311	-0.3875	0.0016	-0.0002	0.0000	0.0000		
557	1	7	0.902	1.00	11.19	0.0995	1.02	-0.04	0.05	-0.01	0.0019	0.1258	0.7148
		8	0.0114	0.0002	0.0014	0.0995	0.6478	0.0135	0.0003	0.0019	0.0014	0.0746	0.6755
		9	0.0013	0.0003	0.0002	1.3333	0.9570	0.0104	0.0001	0.0014	0.0014		
557	2	7	0.902	1.01	22.49	0.1253	1.02	-0.04	0.05	-0.01	0.0013	0.1303	0.7040
		8	0.0126	0.0003	0.0017	0.1253	0.6884	0.0119	0.0003	0.0013	0.0013	0.0813	0.6686
		9	0.0038	0.0003	0.0006	0.5130	0.8897	0.0100	0.0001	0.0013	0.0013	0.0960	0.7220
557	3	7	0.902	1.01	33.69	0.1002	1.02	-0.04	0.06	-0.01	0.0013	0.1702	0.6477
		8	0.0141	0.0002	0.0018	0.1002	0.6678	0.0093	0.0003	0.0013	0.0013	0.2042	0.6745
		9	0.0067	0.0003	0.0008	0.2726	0.6269	0.0089	0.0001	0.0011	0.0009	0.1225	0.6392
557	4	7	0.902	1.01	44.99	0.0983	1.02	-0.05	0.06	-0.01	0.0009	0.2042	0.6745
		8	0.0146	0.0002	0.0019	0.0983	0.6824	0.0068	0.0002	0.0009	0.0009	0.1636	0.6077
		9	0.0075	0.0004	0.0011	0.3168	0.7606	0.0073	0.0001	0.0009	0.0006	0.0888	0.5905
557	5	7	0.902	1.01	56.19	0.0898	1.02	-0.05	0.07	-0.01	0.0005	0.1636	0.6077
		8	0.0149	0.0002	0.0020	0.0898	0.6761	0.0047	0.0001	0.0005	0.0005	0.0021	0.5049
		9	0.0088	0.0005	0.0013	0.2916	0.7385	0.0057	0.0001	0.0006	0.0004	-0.0130	0.4995
557	6	7	0.902	1.01	67.49	0.0964	1.02	-0.05	0.06	-0.02	0.0002	-0.0021	0.5049
		8	0.0147	0.0002	0.0020	0.0964	0.6881	0.0029	0.0000	0.0002	0.0002	0.0021	0.4995
		9	0.0095	0.0005	0.0014	0.2906	0.7679	0.0046	-0.0000	0.0004	0.0004	-0.0130	0.4995
557	7	7	0.902	1.00	78.69	0.1030	1.02	-0.06	0.06	-0.01	0.0000	-0.4850	0.3564
		8	0.0146	0.0003	0.0020	0.1030	0.6966	0.0016	0.0001	0.0000	0.0000	0.4850	0.3564
		9	0.0104	0.0005	0.0014	0.2603	0.7111	0.0033	-0.0001	0.0002	0.0002	-0.1939	0.3564

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																																		
557	8	7	0.902	1.01	89.99	0.0719	1.02	-0.006	0.06	-0.0002	-0.0002	4.1293	4.3627	0.0154	0.0002	0.0020	0.2935	0.6671	-0.0002	-0.0002	-0.0000	-0.0000	0.0099	0.0005	0.0015	0.0719	0.6671	0.7809	0.0017	-0.0002	-0.0002	-0.0000	-0.0000	0.0002	4.1293	4.3627	
558	1	7	0.903	3.07	11.20	0.1005	1.02	-0.0349	0.06	0.0008	0.0046	0.1243	0.6712	0.0149	0.0002	0.0019	0.4377	0.6481	0.0008	0.0006	0.0041	0.0041	0.0968	0.0056	0.0004	0.0007	0.1005	0.6481	0.6988	0.0313	0.0006	0.0006	0.0041	0.0041	0.1243	0.6712	0.6637
558	2	7	0.899	5.07	22.50	0.0854	1.03	-0.02	0.07	0.0007	0.0000	0.1251	0.6674	0.0210	0.0003	0.0027	0.3091	0.6505	0.0007	0.0005	0.0038	0.0038	0.0947	0.0118	0.0007	0.0016	0.0854	0.6505	0.6813	0.0295	0.0005	0.0005	0.0038	0.0038	0.1251	0.6674	0.6534
558	3	7	0.900	3.07	33.69	0.1048	1.04	-0.02	0.08	0.0006	0.0034	0.1200	0.6635	0.0259	0.0005	0.0035	0.2798	0.6745	0.0006	0.0005	0.0034	0.0034	0.0999	0.0171	0.0009	0.0024	0.1048	0.6745	0.7020	0.0264	0.0005	0.0005	0.0034	0.0034	0.1200	0.6635	0.6527
558	4	7	0.901	3.07	44.99	0.0992	1.04	-0.03	0.09	0.0004	0.0029	0.1163	0.6550	0.0312	0.0006	0.0042	0.2435	0.6715	0.0004	0.0004	0.0029	0.0029	0.1099	0.0227	0.0011	0.0031	0.0992	0.6715	0.7026	0.0220	0.0004	0.0004	0.0029	0.0029	0.1163	0.6550	0.6617
558	5	7	0.902	3.08	56.19	0.1019	1.05	-0.04	0.10	0.0003	0.0021	0.1318	0.6495	0.0350	0.0007	0.0047	0.1845	0.6763	0.0003	0.0003	0.0021	0.0021	0.0942	0.0292	0.0010	0.0036	0.1019	0.6763	0.6152	0.0168	0.0003	0.0003	0.0021	0.0021	0.1318	0.6495	0.6495
558	6	7	0.900	3.08	67.49	0.0925	1.05	-0.05	0.11	0.0002	0.0014	0.1677	0.6346	0.0379	0.0007	0.0050	0.2081	0.6654	0.0002	0.0002	0.0014	0.0014	0.0906	0.0305	0.0012	0.0043	0.0925	0.6654	0.7050	0.0111	0.0002	0.0002	0.0014	0.0014	0.1677	0.6346	0.6346
558	7	7	0.900	3.08	78.69	0.0958	1.05	-0.05	0.11	0.0000	0.0006	0.0741	0.5873	0.0387	0.0007	0.0052	0.1891	0.6805	0.0000	0.0000	0.0006	0.0006	0.0741	0.0324	0.0012	0.0044	0.0958	0.6805	0.6805	0.0059	0.0000	0.0000	0.0006	0.0006	0.0741	0.5873	0.5873
558	8	7	0.901	3.07	89.99	0.1124	1.05	-0.06	0.11	0.0003	0.0000	4.1189	3.0197	0.0375	0.0008	0.0052	0.1751	0.6507	0.0003	0.0003	0.0000	0.0000	0.0127	0.0328	0.0011	0.0042	0.1124	0.6507	0.6507	0.0014	0.0003	0.0003	0.0000	0.0000	4.1189	3.0197	0.0356
559	1	7	0.901	6.21	11.20	0.0798	1.03	0.02	0.08	0.0016	0.0081	0.1292	0.6343	0.0194	0.0003	0.0024	0.3672	0.6232	0.0016	0.0015	0.0081	0.0081	0.1170	0.0117	0.0008	0.0016	0.0798	0.6232	0.6955	0.0649	0.0015	0.0015	0.0081	0.0081	0.1292	0.6343	0.6279
559	2	7	0.900	6.20	22.50	0.0915	1.04	0.02	0.10	0.0015	0.0077	0.1240	0.6428	0.0314	0.0005	0.0040	0.2794	0.6485	0.0015	0.0015	0.0077	0.0077	0.1298	0.0243	0.0015	0.0031	0.0915	0.6485	0.6601	0.0015	0.0015	0.0077	0.0077	0.1240	0.6428	0.6433	

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key															
559	3	7	0.901	6.20	33.70	0.0972	1.06	0.01	0.12	0.03	0.1197	0.6555						
		8	0.0433	0.0008	0.0057	0.2452	0.6632	0.0540	0.0012	0.0070	0.1423	0.6474						
		9	0.0363	0.0017	0.0047	0.0841	0.6576	0.0528	0.0015	0.0068								
559	4	7	0.899	6.20	44.99	0.2361	1.07	-0.00	0.14	0.072	0.1255	0.6538						
		8	0.0557	0.0009	0.0072	0.0841	0.6543	0.0438	0.0011	0.0057	0.1531	0.6499						
		9	0.0451	0.0021	0.0058	0.2361	0.6492	0.0431	0.0013	0.0056								
559	5	7	0.900	6.20	56.19	0.0922	1.08	-0.01	0.14	0.01	0.1369	0.6416						
		8	0.0637	0.0011	0.0082	0.2243	0.6472	0.0321	0.0008	0.0041	0.1665	0.6428						
		9	0.0517	0.0023	0.0063	0.2243	0.6143	0.0315	0.0010	0.0040								
559	6	7	0.902	6.20	67.49	0.1005	1.09	-0.03	0.15	0.00	0.1429	0.6137						
		8	0.0684	0.0013	0.0087	0.2075	0.6374	0.0207	0.0005	0.0025	0.1486	0.6309						
		9	0.0568	0.0023	0.0068	0.2075	0.6062	0.0205	0.0006	0.0025								
559	7	7	0.901	6.21	78.69	0.1089	1.10	-0.04	0.15	0.00	0.1677	0.6059						
		8	0.0707	0.0015	0.0088	0.1904	0.6257	0.0093	0.0003	0.0011	0.0870	0.5787						
		9	0.0602	0.0022	0.0072	0.1904	0.6029	0.0091	0.0001	0.0010								
559	8	7	0.902	6.21	89.99	0.1158	1.10	-0.06	0.15	-0.03	-2.1439	-1.5942						
		8	0.0704	0.0016	0.0087	0.1847	0.6204	0.0003	-0.0003	-0.0002	69.3965	54.0615						
		9	0.0598	0.0022	0.0074	0.1847	0.6248	-0.0000	-0.0003	-0.0002								
560	1	7	0.899	9.32	11.20	0.0674	1.03	0.05	0.09	0.07	0.1264	0.6076						
		8	0.0247	0.0003	0.0031	0.3580	0.6312	0.0905	0.0022	0.0110	0.1214	0.5902						
		9	0.0156	0.0011	0.0023	0.3580	0.7365	0.0870	0.0021	0.0102								
560	2	7	0.900	9.33	22.50	0.0863	1.06	0.05	0.12	0.07	0.1146	0.6197						
		8	0.0424	0.0007	0.0055	0.2773	0.6550	0.0883	0.0020	0.0109	0.1364	0.5911						
		9	0.0342	0.0018	0.0045	0.2773	0.6661	0.0809	0.0022	0.0095								
560	3	7	0.899	9.32	33.70	0.0731	1.08	0.04	0.15	0.06	0.1049	0.6355						
		8	0.0615	0.0009	0.0081	0.2578	0.6621	0.0811	0.0017	0.0103	0.1540	0.5952						
		9	0.0486	0.0025	0.0060	0.2578	0.6250	0.0728	0.0022	0.0086								
560	4	7	0.900	9.32	44.99	0.0785	1.10	0.02	0.16	0.05	0.1026	0.6498						
		8	0.0771	0.0012	0.0099	0.2245	0.6472	0.0678	0.0013	0.0088	0.1735	0.6031						
		9	0.0614	0.0027	0.0073	0.2245	0.5948	0.0617	0.0021	0.0074								
560	5	7	0.901	9.31	56.19	0.0846	1.11	0.00	0.18	0.04	0.1154	0.6520						
		8	0.0880	0.0014	0.0110	0.2101	0.6273	0.0500	0.0011	0.0065	0.1786	0.6308						
		9	0.0701	0.0029	0.0084	0.2101	0.6045	0.0491	0.0017	0.0062								

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key														
560	6	7	0.902	9.33	67.49	0.0971	0.1852	1.12	0.6116	0.5890	-0.01	0.0318	0.0008	0.19	0.0039	0.1312	0.6276
		8	0.0928	0.0018	0.0113	0.0113	0.0321	0.5988	0.5988	0.0003	0.0003	0.0004	0.0017	0.18	0.0017	0.1311	0.5798
		9	0.0779	0.0028	0.0091	0.1746	0.0145	0.6001	0.6001	0.0004	0.0004	0.0017	0.0017	0.0004	0.0017	0.1597	0.6110
560	7	7	0.901	9.33	78.69	0.1058	0.1746	1.12	0.5988	0.6001	-0.03	0.0146	0.0003	0.18	0.0017	0.1311	0.5798
		8	0.0947	0.0020	0.0113	0.1746	0.0145	0.6001	0.6001	0.0004	0.0004	0.0017	0.0017	0.0004	0.0017	0.1597	0.6110
		9	0.0822	0.0028	0.0098	0.1746	0.0145	0.6001	0.6001	0.0004	0.0004	0.0017	0.0017	0.0004	0.0017	0.1597	0.6110
560	8	7	0.901	9.33	89.99	0.1206	0.1551	1.12	0.5988	0.5882	-0.06	0.0001	0.0001	0.19	0.0002	3.0857	5.0515
		8	0.0925	0.0022	0.0110	0.1551	0.0001	0.5882	0.5882	0.0001	0.0001	0.0002	0.0002	0.19	0.0002	8.4825	8.7621
		9	0.0860	0.0026	0.0101	0.1551	0.0001	0.5882	0.5882	0.0001	0.0001	0.0002	0.0002	0.19	0.0002	8.4825	8.7621
561	1	7	0.901	12.43	11.20	0.0784	0.3000	1.04	0.6370	0.6243	0.07	0.1109	0.0024	0.10	0.0024	0.1107	0.5935
		8	0.0288	0.0004	0.0036	0.3000	0.0001	0.6243	0.6243	0.0001	0.0001	0.0021	0.0021	0.10	0.0021	0.1107	0.5935
		9	0.0213	0.0012	0.0026	0.3000	0.0001	0.6243	0.6243	0.0001	0.0001	0.0021	0.0021	0.10	0.0021	0.1107	0.5935
561	2	7	0.901	12.43	22.50	0.0818	0.2719	1.07	0.6696	0.6062	0.07	0.1084	0.0022	0.14	0.0022	0.1048	0.6024
		8	0.0521	0.0008	0.0059	0.2719	0.0001	0.6062	0.6062	0.0001	0.0001	0.0022	0.0022	0.14	0.0022	0.1155	0.5718
		9	0.0432	0.0023	0.0052	0.2719	0.0001	0.6062	0.6062	0.0001	0.0001	0.0022	0.0022	0.14	0.0022	0.1155	0.5718
561	3	7	0.902	12.42	33.70	0.0620	0.2436	1.10	0.6587	0.5895	0.06	0.1023	0.0023	0.17	0.0023	0.1001	0.6198
		8	0.0773	0.0009	0.0101	0.2436	0.0001	0.5895	0.5895	0.0001	0.0001	0.0023	0.0023	0.17	0.0023	0.1340	0.5734
		9	0.0609	0.0029	0.0071	0.2436	0.0001	0.5895	0.5895	0.0001	0.0001	0.0023	0.0023	0.17	0.0023	0.1340	0.5734
561	4	7	0.900	12.42	45.00	0.0680	0.2022	1.12	0.6355	0.5755	0.04	0.0881	0.0024	0.18	0.0024	0.0896	0.6403
		8	0.0975	0.0013	0.0124	0.2022	0.0001	0.5755	0.5755	0.0001	0.0001	0.0024	0.0024	0.18	0.0024	0.1602	0.5813
		9	0.0761	0.0030	0.0087	0.2022	0.0001	0.5755	0.5755	0.0001	0.0001	0.0024	0.0024	0.18	0.0024	0.1602	0.5813
561	5	7	0.901	12.42	56.19	0.0848	0.1735	1.13	0.6111	0.5682	0.01	0.0664	0.0023	0.19	0.0023	0.0922	0.6490
		8	0.1072	0.0018	0.0131	0.1735	0.0001	0.5682	0.5682	0.0001	0.0001	0.0023	0.0023	0.19	0.0023	0.1887	0.5966
		9	0.0866	0.0030	0.0098	0.1735	0.0001	0.5682	0.5682	0.0001	0.0001	0.0023	0.0023	0.19	0.0023	0.1887	0.5966
561	6	7	0.901	12.42	67.48	0.0898	0.1499	1.14	0.5993	0.5722	-0.00	0.0419	0.0017	0.20	0.0017	0.1092	0.6383
		8	0.1128	0.0020	0.0135	0.1499	0.0001	0.5722	0.5722	0.0001	0.0001	0.0017	0.0017	0.20	0.0017	0.1988	0.6373
		9	0.0960	0.0028	0.0109	0.1499	0.0001	0.5722	0.5722	0.0001	0.0001	0.0017	0.0017	0.20	0.0017	0.1988	0.6373
561	7	7	0.900	12.43	78.69	0.0941	0.1355	1.15	0.5891	0.5798	-0.03	0.0186	0.0007	0.21	0.0007	0.1303	0.6225
		8	0.1147	0.0021	0.0135	0.1355	0.0001	0.5798	0.5798	0.0001	0.0001	0.0007	0.0007	0.21	0.0007	0.1863	0.5856
		9	0.1034	0.0028	0.0119	0.1355	0.0001	0.5798	0.5798	0.0001	0.0001	0.0007	0.0007	0.21	0.0007	0.1863	0.5856
561	8	7	0.900	12.43	90.00	0.1010	0.1259	1.15	0.5863	0.5760	-0.06	0.0002	0.0001	0.21	0.0002	2.9316	4.2939
		8	0.1134	0.0022	0.0133	0.1259	0.0001	0.5760	0.5760	0.0001	0.0001	0.0002	0.0002	0.21	0.0002	-1.0534	-9.6303
		9	0.1084	0.0027	0.0124	0.1259	0.0001	0.5760	0.5760	0.0001	0.0001	0.0002	0.0002	0.21	0.0002	-1.0534	-9.6303

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key														
562	3	7 8 9	0.898 0.0146 -0.0053	-3.12 0.0002 -0.0000	11.19 0.0019 -0.0005	0.0888 0.0115	3.03 0.5527	-0.10 -0.0269 -0.0286	0.03 -0.0007 -0.0007	-0.05 -0.0036 -0.0036	0.1441 0.1329	0.6725 0.6743					
562	4	7 8 9	0.899 0.0079 -0.0113	-3.11 0.0001 -0.0000	22.49 0.0011 -0.0014	0.0840 0.0090	3.02 0.7203 0.6356	-0.10 -0.0250 -0.0269	0.03 -0.0007 -0.0006	-0.05 -0.0034 -0.0036	0.1513 0.1279	0.6813 0.6765					
562	5	7 8 9	0.898 0.0022 -0.0165	-3.12 -0.0000 -0.0001	33.69 0.0005 -0.0021	-0.0680 0.0460	3.01 1.1470 0.6421	-0.09 -0.0220 -0.0238	0.02 -0.0006 -0.0006	-0.04 -0.0029 -0.0032	0.1575 0.1278	0.6773 0.6796					
562	6	7 8 9	0.899 -0.0021 -0.0221	-3.11 -0.0002 -0.0002	44.99 -0.0000 -0.0026	0.5237 0.0487	3.01 0.0642 0.6017	-0.08 -0.0182 -0.0194	0.01 -0.0005 -0.0004	-0.04 -0.0025 -0.0026	0.1620 0.1261	0.7012 0.6831					
562	7	7 8 9	0.898 -0.0047 -0.0258	-3.11 -0.0004 -0.0003	56.19 -0.0003 -0.0032	0.4347 0.0692	3.00 0.4018 0.6361	-0.09 -0.0133 -0.0142	0.01 -0.0005 -0.0003	-0.04 -0.0019 -0.0020	0.1888 0.1305	0.7366 0.7060					
562	8	7 8 9	0.899 -0.0074 -0.0290	-3.12 -0.0004 -0.0004	67.49 -0.0007 -0.0037	0.3047 0.0791	3.00 0.4843 0.6509	-0.08 -0.0082 -0.0085	0.01 -0.0004 -0.0002	-0.03 -0.0013 -0.0013	0.2437 0.1672	0.8074 0.7767					
562	9	7 8 9	0.898 -0.0092 -0.0324	-3.11 -0.0004 -0.0004	78.69 -0.0010 -0.0040	0.2546 0.0696	3.00 0.5483 0.6185	-0.07 -0.0027 -0.0030	0.00 -0.0002 -0.0002	-0.03 -0.0006 -0.0006	0.4790 0.3993	1.1340 0.9992					
562	10	7 8 9	0.899 -0.0104 -0.0323	-3.11 -0.0004 -0.0005	89.99 -0.0011 -0.0042	0.2032 0.0881	3.00 0.5336 0.6589	-0.07 0.0026 0.0016	0.01 -0.0000 -0.0001	-0.03 0.0002 0.0000	-0.1533 -0.4348	0.4567 0.2265					
563	1	7 8 9	0.898 0.0242 -0.0002	-0.01 0.0002 0.0002	11.19 0.0032 0.0000	0.0563 -5.6125	3.04 0.6759 -1.4569	-0.06 0.0040 0.0020	0.04 -0.0001 -0.0000	-0.02 0.0005 0.0002	0.1670 -0.2218	0.7286 0.5194					
563	2	7 8 9	0.899 0.0233 0.0011	-0.00 0.0003 0.0002	22.49 0.0032 0.0000	0.0763 0.9666	3.04 0.7053 0.0276	-0.06 0.0037 0.0019	0.04 -0.0001 -0.0000	-0.02 0.0005 0.0001	0.1388 -0.2360	0.6985 0.4964					
563	3	7 8 9	0.899 0.0226 0.0005	-0.00 0.0002 0.0002	33.69 0.0031 0.0000	0.0595 2.5911	3.03 0.6879 0.7407	-0.06 0.0033 0.0018	0.04 -0.0000 -0.0001	-0.03 0.0004 0.0001	0.0657 -0.2825	0.6555 0.4922					

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key												
563	4	7	0.900	-0.0002	44.77	0.0552	3.04	-0.06	0.04	0.0000	0.04	-0.0003	-0.0280	0.6037
		8	0.0212	0.0002	0.0000	11.0920	0.6723	0.0018	-0.0001	-0.0000	-0.0000	0.0001	-0.3163	0.4591
		9	0.0001	0.0002	0.0000	0.0000	3.4636	0.0018	-0.0001	0.0000	0.0000	0.0001	0.0000	0.0000
563	5	7	0.898	-0.0002	56.19	0.0621	3.04	-0.06	0.04	0.0000	0.04	-0.0001	-0.2114	0.5164
		8	0.0200	0.0002	0.0000	-22.2461	0.6860	0.0017	-0.0001	-0.0000	-0.0000	0.0001	-0.4067	0.3921
		9	-0.0000	0.0002	0.0000	0.0000	-7.4100	0.0017	-0.0001	0.0000	0.0000	0.0001	0.0000	0.0000
563	6	7	0.898	-0.0002	67.49	0.0591	3.04	-0.06	0.04	0.0000	0.04	-0.0000	-0.5710	0.0847
		8	0.0194	0.0002	0.0026	-5.3726	0.6842	0.0011	-0.0001	-0.0001	-0.0001	0.0000	-0.5208	0.3131
		9	-0.0002	0.0002	0.0001	0.0000	-1.9410	0.0017	-0.0001	0.0000	0.0000	0.0001	0.0000	0.0000
563	7	7	0.897	-0.0003	78.69	0.0975	3.04	-0.06	0.04	0.0000	0.04	-0.0001	-3.3722	-1.9161
		8	0.0184	0.0003	0.0027	-6.2421	0.7437	0.0002	-0.0001	-0.0002	-0.0002	0.0000	-0.7000	0.0714
		9	-0.0002	0.0003	0.0000	0.0000	-1.9753	0.0002	-0.0001	0.0000	0.0000	0.0000	0.0000	0.0000
563	8	7	0.898	-0.0001	89.99	0.0406	3.03	-0.06	0.04	0.0000	0.04	-0.0001	-4.3900	-3.2277
		8	0.0202	0.0001	0.0026	-1.9452	0.6585	0.0002	-0.0001	-0.0002	-0.0002	0.0000	-1.0706	-0.1640
		9	-0.0008	0.0003	0.0001	0.0000	-0.8761	0.0001	-0.0001	0.0000	0.0000	0.0000	0.0000	0.0000
564	1	7	0.898	1.02	11.20	0.0743	3.04	-0.04	0.04	0.0000	0.04	-0.0001	0.1195	0.7290
		8	0.0266	0.0003	0.0035	0.8986	0.6757	0.0099	0.0003	0.0003	0.0003	0.0003	0.0921	0.6872
		9	0.0019	0.0003	0.0002	0.0000	0.6769	0.0099	0.0001	0.0001	0.0001	0.0001	0.0000	0.0000
564	2	7	0.896	1.02	22.50	0.0740	3.04	-0.04	0.05	0.0000	0.05	-0.0001	0.1261	0.7258
		8	0.0282	0.0004	0.0038	0.4620	0.6786	0.0094	0.0001	0.0001	0.0001	0.0001	0.1043	0.6979
		9	0.0042	0.0003	0.0006	0.0000	0.7904	0.0094	0.0001	0.0001	0.0001	0.0001	0.0000	0.0000
564	3	7	0.897	1.02	33.69	0.0831	3.05	-0.05	0.05	0.0000	0.05	-0.0001	0.1486	0.7103
		8	0.0290	0.0004	0.0040	0.3530	0.6929	0.0099	0.0002	0.0002	0.0002	0.0001	0.1283	0.6722
		9	0.0061	0.0004	0.0009	0.0000	0.7710	0.0082	0.0002	0.0002	0.0002	0.0001	0.0000	0.0000
564	4	7	0.897	1.02	44.99	0.0747	3.04	-0.06	0.06	0.0000	0.06	-0.0001	0.2092	0.7056
		8	0.0298	0.0004	0.0041	0.3035	0.6923	0.0068	0.0001	0.0001	0.0001	0.0001	0.1211	0.6444
		9	0.0078	0.0004	0.0011	0.0000	0.7367	0.0068	0.0001	0.0001	0.0001	0.0001	0.0000	0.0000
564	5	7	0.896	1.02	56.19	0.0869	3.05	-0.05	0.05	0.0000	0.05	-0.0001	0.1919	0.6835
		8	0.0295	0.0005	0.0041	0.3082	0.7090	0.0046	0.0001	0.0001	0.0001	0.0001	0.0951	0.6158
		9	0.0086	0.0005	0.0013	0.0000	0.7831	0.0054	0.0001	0.0001	0.0001	0.0001	0.0000	0.0000
564	6	7	0.898	1.02	67.49	0.0756	3.05	-0.06	0.06	0.0000	0.06	-0.0001	0.0115	0.5928
		8	0.0301	0.0004	0.0041	0.3032	0.6889	0.0031	0.0000	0.0000	0.0000	0.0000	-0.0061	0.5597
		9	0.0095	0.0005	0.0015	0.0000	0.7937	0.0042	-0.0000	-0.0000	-0.0000	0.0000	0.0000	0.0000

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																																								
564	1	0.899	1.03	78.69	0.0751	3.05	-0.018	0.06	-0.002	-0.4210	0.2106	0.0301	0.0004	0.0041	0.0751	0.6888	0.0018	-0.0001	0.0000	-0.0002	-0.4210	0.2106	0.0101	0.0005	0.0015	0.2844	0.7706	0.0030	-0.0001	0.0002	0.0002	-0.2377	0.3710									
564	8	0.899	1.03	89.99	0.0788	3.05	-0.007	0.06	-0.003	11.9001	10.4780	0.0302	0.0004	0.0042	0.2430	0.6944	-0.0001	-0.0002	-0.0000	-0.0002	11.9001	10.4780	0.0109	0.0005	0.0015	0.2430	0.6946	0.0012	-0.0002	-0.0000	-1.1397	-0.2266	0.0109	0.0005	0.0015	0.2430	0.6946	0.0012	-0.0002	-0.0000	-1.1397	-0.2266
565	1	0.898	3.09	11.20	0.0966	3.05	-0.01	0.05	0.01	0.1247	0.6754	0.0291	0.0005	0.0038	0.0966	0.6663	0.0351	0.0008	0.0040	0.0040	0.1247	0.6754	0.0057	0.0004	0.0007	0.4252	0.6868	0.0308	0.0006	0.0006	0.0995	0.6553	0.0057	0.0004	0.0006	0.0995	0.6553					
565	2	0.899	3.09	22.50	0.1040	3.05	-0.02	0.06	-0.00	0.1257	0.6744	0.0348	0.0007	0.0047	0.1040	0.6787	0.0308	0.0007	0.0041	0.0037	0.1257	0.6744	0.0119	0.0007	0.0016	0.3037	0.6756	0.0285	0.0006	0.0006	0.1052	0.6568	0.0119	0.0007	0.0016	0.1052	0.6568					
565	3	0.898	3.09	33.70	0.1009	3.06	-0.03	0.07	-0.00	0.1183	0.6654	0.0407	0.0008	0.0054	0.1009	0.6747	0.0263	0.0006	0.0035	0.0035	0.1183	0.6654	0.0175	0.0009	0.0024	0.2705	0.7055	0.0255	0.0005	0.0005	0.1147	0.6573	0.0175	0.0009	0.0024	0.1147	0.6573					
565	4	0.898	3.09	45.00	0.1023	3.07	-0.03	0.08	-0.00	0.1158	0.6659	0.0458	0.0009	0.0061	0.1023	0.6701	0.0208	0.0004	0.0027	0.0027	0.1158	0.6659	0.0239	0.0010	0.0031	0.2169	0.6512	0.0213	0.0004	0.0004	0.1143	0.6515	0.0239	0.0010	0.0031	0.1143	0.6515					
565	5	0.898	3.09	56.19	0.1082	3.07	-0.04	0.08	-0.01	0.1234	0.6755	0.0493	0.0010	0.0066	0.1082	0.6694	0.0147	0.0003	0.0019	0.0019	0.1234	0.6755	0.0277	0.0011	0.0038	0.2139	0.6920	0.0163	0.0003	0.0003	0.1028	0.6538	0.0277	0.0011	0.0038	0.1028	0.6538					
565	6	0.896	3.10	67.49	0.1117	3.07	-0.05	0.09	-0.01	0.1619	0.6920	0.0513	0.0011	0.0068	0.1117	0.6696	0.0085	0.0002	0.0013	0.0013	0.1619	0.6920	0.0320	0.0011	0.0041	0.1811	0.6476	0.0104	0.0002	0.0002	0.1063	0.6432	0.0320	0.0011	0.0041	0.1063	0.6432					
565	7	0.900	3.10	78.69	0.1142	3.08	-0.06	0.09	-0.02	0.0822	0.6604	0.0516	0.0011	0.0069	0.1142	0.6684	0.0038	0.0000	0.0005	0.0005	0.0822	0.6604	0.0325	0.0012	0.0044	0.1886	0.6831	0.0056	0.0000	0.0000	0.0010	0.5162	0.6604	0.0325	0.0012	0.0044	0.0010	0.5162	0.6604			
565	8	0.899	3.10	89.99	0.1120	3.08	-0.07	0.09	-0.03	3.5659	1.7515	0.0516	0.0011	0.0069	0.1120	0.6714	0.0003	0.0002	0.0001	0.0001	3.5659	1.7515	0.0320	0.0012	0.0045	0.1895	0.7043	0.0010	0.0003	0.0003	0.0000	0.3022	0.6515	0.0320	0.0012	0.0045	0.0000	0.3022	0.6515			
566	1	0.897	6.23	11.20	0.0858	3.05	0.02	0.07	0.04	0.1314	0.6391	0.0332	0.0008	0.0043	0.0858	0.6809	0.0655	0.0017	0.0083	0.0079	0.1314	0.6391	0.0125	0.0008	0.0016	0.3285	0.6547	0.0637	0.0016	0.0016	0.1283	0.6251	0.0125	0.0008	0.0016	0.1283	0.6251					

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd																	
566	2	7	0.899	6.22	22.30	0.0872	3.07	0.01	0.08	0.003	0.1228	0.6486						
		8	0.6460	0.0008	0.0061	0.0672	0.6646	0.0610	0.0014	0.0079	0.1417	0.6409						
		9	0.0248	0.0013	0.0031	0.2685	0.6439	0.0586	0.0016	0.0075								
566	3	7	0.898	6.22	33.70	0.0842	3.08	0.00	0.11	0.003	0.1167	0.6607						
		8	0.586	0.0009	0.0077	0.2325	0.6626	0.0544	0.0012	0.0071	0.1469	0.6466						
		9	0.0374	0.0017	0.0048		0.6540	0.0522	0.0015	0.0067								
566	4	7	0.898	6.21	44.99	0.0825	3.10	-0.00	0.12	0.002	0.1253	0.6631						
		8	0.692	0.0011	0.0088	0.2307	0.6390	0.0439	0.0011	0.0058	0.1598	0.6496						
		9	0.0458	0.0021	0.0059		0.6509	0.0422	0.0013	0.0054								
566	5	7	0.899	6.22	56.19	0.0953	3.11	-0.01	0.14	0.001	0.1362	0.6488						
		8	0.754	0.0014	0.0094	0.2234	0.6282	0.0320	0.0008	0.0041	0.1689	0.6432						
		9	0.0520	0.0023	0.0065		0.6275	0.0306	0.0010	0.0039								
566	6	7	0.898	6.23	67.49	0.1126	3.12	-0.03	0.14	-0.000	0.1383	0.6223						
		8	0.785	0.0017	0.0098	0.1950	0.6255	0.0208	0.0005	0.0025	0.1573	0.6424						
		9	0.0585	0.0022	0.0069		0.6237	0.0194	0.0006	0.0025								
566	7	7	0.898	6.23	78.69	0.1194	3.12	-0.05	0.14	-0.001	0.1508	0.6163						
		8	0.800	0.0019	0.0099	0.1953	0.6198	0.0096	0.0002	0.0011	0.1045	0.6000						
		9	0.0598	0.0023	0.0075		0.6285	0.0083	0.0001	0.0009								
566	8	7	0.899	6.22	89.99	0.1261	3.12	-0.06	0.15	-0.003	-1.8888	-0.8980						
		8	0.793	0.0020	0.0097	0.1777	0.6126	0.0004	-0.0003	0.0000	-2.2082	-2.2457						
		9	0.0611	0.0021	0.0075		0.6202	-0.0007	-0.0003	0.0003								
567	1	7	0.899	9.34	11.20	0.0666	3.06	0.06	0.07	0.006	0.1245	0.6116						
		8	0.390	0.0005	0.0050	0.2878	0.6492	0.0912	0.0022	0.0111	0.1254	0.5902						
		9	0.0178	0.0010	0.0022		0.6258	0.0859	0.0021	0.0101								
567	2	7	0.899	9.33	22.50	0.0699	3.08	0.05	0.11	0.006	0.1122	0.6251						
		8	0.579	0.0008	0.0076	0.2581	0.6619	0.0689	0.0019	0.0111	0.1424	0.5941						
		9	0.0355	0.0018	0.0045		0.6390	0.0795	0.0022	0.0094								
567	3	7	0.898	9.32	33.70	0.0719	3.10	0.04	0.13	0.005	0.1044	0.6396						
		8	0.753	0.0010	0.0098	0.2538	0.6517	0.0610	0.0016	0.0103	0.1588	0.5949						
		9	0.0490	0.0024	0.0061		0.6281	0.0719	0.0022	0.0085								
567	4	7	0.898	9.33	45.00	0.0777	3.12	0.02	0.16	0.004	0.1007	0.6531						
		8	0.895	0.0013	0.0113	0.2263	0.6343	0.0678	0.0013	0.0088	0.1782	0.6028						
		9	0.0613	0.0027	0.0074		0.6086	0.0609	0.0021	0.0073								

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																	
567	5	7	0.899	9.33	56.19	0.0955	3.13	0.00	0.17	0.02	0.1141	0.6517	0.0707	0.0029	0.0018	0.0029	0.0017	0.0065	0.1840	0.6341
567	6	5	0.899	9.33	67.49	0.1025	3.14	-0.02	0.18	0.01	0.1286	0.6307	0.1000	0.0020	0.0120	0.0091	0.0012	0.0039	0.1931	0.6364
567	7	7	0.899	9.34	78.69	0.1101	3.14	-0.04	0.18	-0.00	0.1310	0.5890	0.1011	0.0022	0.0120	0.0098	0.0003	0.0016	0.1561	0.6102
567	8	5	0.897	9.34	89.99	0.1157	3.15	-0.06	0.18	-0.03	3.5126	4.0046	0.0858	0.0027	0.0103	0.0103	-0.0002	-0.0003	2.0977	2.5622
568	1	7	0.898	12.46	11.20	0.0694	3.07	0.07	0.08	0.07	0.1095	0.5956	0.0429	0.0005	0.0056	0.0123	0.0021	0.0133	0.1024	0.5714
568	2	5	0.896	12.46	22.50	0.0602	3.09	0.07	0.13	0.07	0.1048	0.6063	0.0678	0.0008	0.0090	0.0054	0.0023	0.0111	0.1186	0.5713
568	3	7	0.898	12.46	33.70	0.0609	3.12	0.06	0.17	0.06	0.0995	0.6235	0.0614	0.0030	0.0118	0.0073	0.0024	0.0101	0.1369	0.5721
568	4	5	0.898	12.46	45.00	0.0780	3.14	0.04	0.18	0.06	0.0877	0.6427	0.0766	0.0031	0.0089	0.0089	0.0025	0.0089	0.1631	0.5820
568	5	7	0.899	12.47	56.19	0.0849	3.16	0.02	0.19	0.05	0.0916	0.6515	0.1153	0.0019	0.0138	0.0099	0.0012	0.0086	0.1922	0.5959
568	6	5	0.898	12.47	67.48	0.0903	3.16	-0.00	0.20	0.03	0.1081	0.6421	0.0974	0.0028	0.0110	0.0028	0.0017	0.0055	0.2037	0.6387
568	7	7	0.899	12.47	78.68	0.0929	3.16	-0.04	0.21	-0.00	0.1262	0.5983	0.1212	0.0022	0.0141	0.0121	0.0007	0.0025	0.1869	0.6305

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key															
568	8	9	0.899	12.47	90.00	0.0968	3.16	-0.006	0.21	-0.0001	-0.0002	0.16099	1.8365					
			0.1176	0.0022	0.0137	0.1231	0.5846	-0.0004	-0.0002	-0.0001	-0.0002	2.8123	2.7933					
			0.1097	0.0027	0.0126	0.1156	0.5754	-0.0004	-0.0002	-0.0002	-0.0002	0.1610	0.6882					
569	1	9	0.898	-3.10	11.19	0.1156	6.06	-0.10	0.04	-0.0008	-0.0004	0.1386	0.6914					
			0.0364	0.0008	0.0049	-0.0434	0.6858	-0.0259	-0.0008	-0.0008	-0.0004	0.1625	0.6957					
			-0.0048	0.0000	-0.0005	0.1077	0.5883	-0.0294	-0.0008	-0.0007	-0.0003	0.1358	0.6957					
569	2	9	0.300	-3.11	22.49	0.1077	6.05	-0.09	0.04	-0.0007	-0.0003	0.1730	0.6971					
			0.0294	0.0006	0.0040	0.0044	0.6929	-0.0243	-0.0007	-0.0003	-0.0003	0.1353	0.6988					
			-0.0105	-0.0000	-0.0014	0.0984	0.6716	-0.0276	-0.0006	-0.0006	-0.0003	0.1773	0.7042					
569	3	9	0.899	-3.11	33.69	0.0984	6.04	-0.09	0.03	-0.0007	-0.0005	0.1530	0.7012					
			0.0231	0.0004	0.0032	0.0331	0.6999	-0.0212	-0.0006	-0.0006	-0.0003	0.2856	0.7836					
			-0.0160	-0.0001	-0.0020	0.0819	0.6446	-0.0246	-0.0005	-0.0005	-0.0002	0.1539	0.7836					
569	4	9	0.900	-3.10	44.99	0.0819	6.03	-0.09	0.01	-0.0008	-0.0004	0.5804	1.1167					
			0.0175	0.0002	0.0024	0.0578	0.6882	-0.0176	-0.0005	-0.0005	-0.0002	0.5309	1.0511					
			-0.0209	-0.0002	-0.0026	0.1644	0.6419	-0.0203	-0.0003	-0.0003	-0.0002	-0.1802	0.5463					
569	5	9	0.899	-3.10	56.19	0.1644	6.03	-0.08	0.01	-0.0008	-0.0004	0.5804	1.1167					
			0.0117	0.0003	0.0017	0.0740	0.7656	-0.0131	-0.0005	-0.0005	-0.0002	0.5309	1.0511					
			-0.0250	-0.0003	-0.0032	0.1479	0.6534	-0.0148	-0.0003	-0.0003	-0.0002	-0.1802	0.5463					
569	6	9	0.898	-3.09	67.49	0.1479	6.03	-0.07	0.00	-0.0004	-0.0001	0.2856	0.7836					
			0.0092	0.0002	0.0014	0.0749	0.7685	-0.0077	-0.0002	-0.0002	-0.0001	0.1539	0.7836					
			-0.0288	-0.0004	-0.0037	0.1716	0.6428	-0.0095	-0.0002	-0.0002	-0.0001	0.2856	0.7836					
569	7	9	0.899	-3.10	78.69	0.1716	6.02	-0.06	0.00	-0.0002	-0.0001	0.5804	1.1167					
			0.0074	0.0002	0.0012	0.0798	0.8446	-0.0024	-0.0002	-0.0002	-0.0001	0.5309	1.0511					
			-0.0311	-0.0004	-0.0040	0.1776	0.6427	-0.0037	-0.0002	-0.0002	-0.0001	-0.1802	0.5463					
569	8	9	0.897	-3.09	89.99	0.1776	6.02	-0.05	0.00	-0.0001	-0.0001	0.1802	0.5463					
			0.0068	0.0002	0.0011	0.0783	0.8562	0.0028	-0.0001	-0.0001	-0.0001	-0.7841	-0.2225					
			-0.0326	-0.0005	-0.0041	0.0961	0.6319	0.0010	-0.0001	-0.0001	-0.0001	0.1443	0.7494					
570	1	9	0.899	-0.01	11.19	0.0961	6.07	-0.06	0.04	-0.0001	-0.0001	0.1443	0.7494					
			0.0452	0.0008	0.0060	2.4236	0.6717	0.0042	0.0001	0.0001	0.0001	-0.5449	0.4096					
			0.0005	0.0002	0.0000	0.1094	0.4488	0.0012	-0.0001	-0.0001	-0.0001	0.0802	0.6702					
570	2	9	0.899	-0.01	22.49	0.1094	6.07	-0.05	0.04	-0.0001	-0.0001	0.0802	0.6702					
			0.0437	0.0009	0.0060	1.8164	0.6944	0.0040	0.0000	0.0000	0.0000	-0.5267	0.5750					
			0.0007	0.0002	0.0001	0.18164	0.7682	0.0013	-0.0001	-0.0001	-0.0001	0.0802	0.6702					

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key →																				
570	3	7	0.896	0.0008	0.0003	33.69	0.0922	6.07	-0.05	0.04	-0.03	0.0409	0.6858										
	6	9	0.0438	0.0004	0.0003	0.0059	3.1868	0.6736	0.0055	0.0000	0.0004	-0.6144	0.3259										
								1.7962	0.0012	-0.0001	0.0000	0.0000											
570	4	7	0.897	0.0008	0.0002	44.99	0.1009	6.07	-0.06	0.04	-0.03	-0.0788	0.5931										
	6	9	0.0422	0.0006	0.0002	0.0058	2.3274	0.6882	0.0029	0.0000	0.0003	-0.6239	0.2171										
								1.0295	0.0012	-0.0001	0.0000	0.0000											
570	5	7	0.897	0.0007	0.0002	56.19	0.0950	6.06	-0.06	0.04	-0.03	-0.2739	0.5304										
	6	9	0.0411	0.0002	0.0002	0.0056	5.8893	0.6910	0.0019	0.0001	0.0002	-0.7921	0.0527										
								2.9454	0.0011	-0.0001	0.0000	0.0000											
570	6	7	0.897	0.0008	0.0002	67.49	0.1007	6.06	-0.06	0.04	-0.03	-0.5971	0.2289										
	6	9	0.0403	0.0002	0.0002	0.0056	15.0446	0.6981	0.0014	0.0001	0.0000	-0.9352	-0.1429										
								8.2069	0.0011	-0.0002	-0.0000	0.0000											
570	7	7	0.897	0.0008	0.0002	78.69	0.1064	6.06	-0.06	0.04	-0.03	-1.3154	-0.2479										
	6	9	0.0400	0.0003	0.0002	0.0056	4.6223	0.7059	0.0008	0.0002	-0.0000	-1.3851	-0.4462										
								2.2937	0.0009	-0.0002	-0.0000	0.0000											
570	8	7	0.897	0.0012	0.0003	89.99	0.0847	6.07	-0.06	0.04	-0.03	-1.7191	-0.4268										
	6	9	0.0412	0.0006	0.0003	0.0056	-3.5060	0.6787	0.0006	0.0002	-0.0001	-2.4901	-1.4391										
								-2.2128	0.0005	-0.0002	-0.0000	0.0000											
571	1	7	0.897	1.02	0.0003	11.20	0.1039	6.07	-0.04	0.05	-0.02	0.1141	0.7221										
	6	9	0.0474	0.0009	0.0003	0.0064	0.9231	0.6768	0.0140	0.0003	0.0020	0.1115	0.6773										
								0.9127	0.0087	0.0001	0.0011	0.1236	0.6685										
571	2	7	0.898	1.03	0.0003	22.50	0.1085	6.08	-0.04	0.04	-0.02	0.1236	0.7228										
	6	9	0.0488	0.0010	0.0003	0.0066	0.4180	0.6772	0.0123	0.0002	0.0011	0.1231	0.6685										
								0.7695	0.0083	0.0002	0.0011	0.1429	0.6677										
571	3	7	0.896	1.03	0.0004	33.70	0.1112	6.08	-0.04	0.05	-0.02	0.1429	0.7192										
	6	9	0.0499	0.0011	0.0004	0.0067	0.3746	0.6756	0.0100	0.0002	0.0009	0.1460	0.6677										
								0.8708	0.0071	0.0002	0.0009	0.1736	0.6073										
571	4	7	0.898	1.03	0.0004	44.99	0.1039	6.07	-0.04	0.05	-0.02	0.1736	0.7033										
	6	9	0.0507	0.0010	0.0004	0.0067	0.3104	0.6875	0.0072	0.0002	0.0007	0.1129	0.6073										
								0.7826	0.0059	0.0001	0.0007	0.1326	0.6566										
571	5	7	0.898	1.03	0.0005	56.19	0.1087	6.08	-0.05	0.05	-0.02	0.1326	0.6566										
	6	9	0.0505	0.0010	0.0005	0.0068	0.2907	0.6745	0.0050	0.0001	0.0006	0.0559	0.5412										
								0.7694	0.0049	0.0000	0.0005	0.1326	0.6566										

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																					
571	6	0.899	1.04	67.49	0.1134	6.07	-0.05	0.05	-0.03	0.0004	-0.0211	0.6299	0.0501	0.0011	0.0068	0.1134	6.07	-0.05	0.05	-0.03	0.0004	-0.0211	0.6299
	6	0.0501	0.0011	0.0068	0.1134	0.6825	0.0032	-0.0000	0.0003	0.0000	-0.0370	0.4973	0.0108	0.0005	0.0015	0.2485	0.7009	0.0036	-0.0000	0.0003	0.0000	-0.0370	0.4973
571	7	0.898	1.04	78.69	0.1195	6.08	-0.06	0.05	-0.03	0.0001	-0.4567	0.2609	0.0497	0.0011	0.0068	0.1195	6.08	-0.06	0.05	-0.03	0.0001	-0.4567	0.2609
	6	0.0497	0.0011	0.0068	0.1195	0.6874	0.0019	-0.0001	0.0001	0.0001	-0.3639	0.2438	0.0111	0.0005	0.0016	0.2609	0.7259	0.0024	-0.0001	0.0001	0.0001	-0.3639	0.2438
571	8	0.898	1.04	89.99	0.1195	6.08	-0.06	0.06	-0.03	0.0001	-24.0136	-14.9026	0.0501	0.0011	0.0068	0.1195	6.08	-0.06	0.06	-0.03	0.0001	-24.0136	-14.9026
	6	0.0501	0.0011	0.0068	0.1195	0.6871	0.0000	-0.0003	-0.0001	-0.0001	-3.0800	-1.5069	0.0104	0.0006	0.0017	0.2905	0.8120	0.0004	-0.0003	-0.0001	-0.0001	-3.0800	-1.5069
572	3	0.902	3.11	11.20	0.0977	6.07	-0.02	0.06	0.00	0.0009	0.1390	0.6027	0.0502	0.0009	0.0068	0.0977	6.07	-0.02	0.06	0.0009	0.1390	0.6027	
	6	0.0502	0.0009	0.0068	0.0977	0.6835	0.0341	0.0009	0.0046	0.0039	0.1020	0.6586	0.0063	0.0004	0.0009	0.3320	0.7299	0.0302	0.0006	0.0039	0.1020	0.6586	
572	4	0.899	3.11	22.50	0.0993	6.08	-0.03	0.07	-0.00	0.0000	0.1424	0.6777	0.0567	0.0011	0.0076	0.0993	6.08	-0.03	0.07	-0.00	0.0000	0.1424	0.6777
	6	0.0567	0.0011	0.0076	0.0993	0.6758	0.0300	0.0008	0.0040	0.0036	0.1085	0.6617	0.0126	0.0006	0.0017	0.2554	0.6973	0.0277	0.0006	0.0036	0.1085	0.6617	
572	5	0.899	3.12	33.70	0.0975	6.09	-0.03	0.09	-0.00	0.0000	0.1468	0.6034	0.0618	0.0012	0.0081	0.0975	6.09	-0.03	0.09	-0.00	0.0000	0.1468	0.6034
	6	0.0618	0.0012	0.0081	0.0975	0.6556	0.0252	0.0007	0.0034	0.0032	0.1251	0.6698	0.0187	0.0008	0.0024	0.2184	0.6622	0.0243	0.0006	0.0032	0.1251	0.6698	
572	6	0.899	3.12	45.00	0.1034	6.10	-0.04	0.09	-0.01	0.0004	0.1405	0.6583	0.0652	0.0013	0.0084	0.1034	6.10	-0.04	0.09	-0.01	0.0004	0.1405	0.6583
	6	0.0652	0.0013	0.0084	0.1034	0.6469	0.0199	0.0005	0.0026	0.0004	0.1185	0.6705	0.0241	0.0009	0.0032	0.2058	0.6733	0.0206	0.0004	0.0027	0.1185	0.6705	
572	7	0.900	3.12	56.20	0.1104	6.10	-0.05	0.10	-0.02	0.0004	0.1672	0.6932	0.0675	0.0014	0.0086	0.1104	6.10	-0.05	0.10	-0.02	0.0004	0.1672	0.6932
	6	0.0675	0.0014	0.0086	0.1104	0.6407	0.0138	0.0004	0.0020	0.0003	0.1042	0.6696	0.0292	0.0010	0.0038	0.1790	0.6539	0.0156	0.0003	0.0020	0.1042	0.6696	
572	8	0.897	3.13	67.49	0.1159	6.10	-0.06	0.11	-0.02	0.0002	0.1709	0.6966	0.0687	0.0015	0.0087	0.1159	6.10	-0.06	0.11	-0.02	0.0002	0.1709	0.6966
	6	0.0687	0.0015	0.0087	0.1159	0.6363	0.0082	0.0002	0.0012	0.0002	0.1052	0.7011	0.0321	0.0011	0.0043	0.1720	0.6687	0.0096	0.0002	0.0012	0.1052	0.7011	
572	9	0.899	3.13	78.69	0.1233	6.10	-0.06	0.10	-0.02	0.0000	0.1052	0.7011	0.0684	0.0016	0.0087	0.1233	6.10	-0.06	0.10	-0.02	0.0000	0.1052	0.7011
	6	0.0684	0.0016	0.0087	0.1233	0.6364	0.0037	0.0000	0.0005	0.0005	-0.0281	0.5074	0.0345	0.0010	0.0044	0.1530	0.6371	0.0051	0.0000	0.0005	-0.0281	0.5074	
572	10	0.897	3.13	89.99	0.1227	6.10	-0.08	0.10	-0.02	0.0003	0.0379	1.4961	0.0683	0.0016	0.0086	0.1227	6.10	-0.08	0.10	-0.02	0.0003	0.0379	1.4961
	6	0.0683	0.0016	0.0086	0.1227	0.6293	-0.0003	-0.0003	-0.0001	-0.0001	-4.3777	-1.0472	0.0333	0.0010	0.0044	0.1619	0.6639	-0.0003	-0.0003	-0.0001	-4.3777	-1.0472	

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																			
573	1	7	0.897	6.25	11.20	0.0694	6.07	0.0657	0.08	0.04	0.1293	0.6539	0.0561	0.0007	0.0016	0.0709	0.0626	0.0016	0.0085	0.1352	0.6328
		9	0.0133	0.0007	0.0016	0.2814	0.6297	0.0626	0.0016	0.0079	0.0016	0.6309	0.0079	0.0007	0.0016	0.6297	0.0626	0.0016	0.0085	0.1352	0.6328
573	2	7	0.898	6.25	22.50	0.0767	6.09	0.0610	0.10	0.04	0.1215	0.6581	0.0678	0.0010	0.0088	0.6496	0.0572	0.0014	0.0080	0.1467	0.6309
		9	0.0250	0.0012	0.0033	0.2588	0.6596	0.0572	0.0016	0.0073	0.0016	0.6309	0.0073	0.0010	0.0088	0.6596	0.0572	0.0016	0.0080	0.1467	0.6309
573	3	7	0.901	6.25	33.70	0.0832	6.10	0.0536	0.12	0.03	0.1179	0.6678	0.0775	0.0012	0.0098	0.6348	0.0503	0.0012	0.0071	0.1556	0.6495
		9	0.0380	0.0016	0.0048	0.2162	0.6387	0.0503	0.0015	0.0065	0.0015	0.6495	0.0065	0.0012	0.0098	0.6387	0.0503	0.0015	0.0071	0.1556	0.6495
573	4	7	0.897	6.25	45.00	0.0912	6.11	-0.00	0.14	0.02	0.1274	0.6652	0.0853	0.0015	0.0105	0.6199	0.0431	0.0010	0.0057	0.1703	0.6525
		9	0.0461	0.0020	0.0060	0.2247	0.6503	0.0404	0.0013	0.0052	0.0013	0.6525	0.0461	0.0020	0.0060	0.6503	0.0404	0.0013	0.0052	0.1703	0.6525
573	5	7	0.898	6.25	56.19	0.1038	6.12	-0.02	0.15	0.00	0.1374	0.6507	0.0892	0.0018	0.0108	0.6069	0.0315	0.0008	0.0041	0.1770	0.6433
		9	0.0537	0.0021	0.0064	0.2042	0.5982	0.0294	0.0010	0.0037	0.0010	0.6433	0.0537	0.0021	0.0064	0.5982	0.0294	0.0010	0.0041	0.1770	0.6433
573	6	7	0.896	6.25	67.49	0.1124	6.13	-0.03	0.16	-0.01	0.1432	0.6327	0.0916	0.0020	0.0109	0.5997	0.0201	0.0005	0.0025	0.1570	0.6492
		9	0.0571	0.0023	0.0072	0.2057	0.6302	0.0186	0.0005	0.0024	0.0005	0.6492	0.0571	0.0023	0.0072	0.6302	0.0186	0.0005	0.0024	0.1570	0.6492
573	7	7	0.899	6.26	78.69	0.1257	6.13	-0.05	0.16	-0.02	0.1658	0.6380	0.0908	0.0022	0.0109	0.6033	0.0090	0.0003	0.0011	0.1658	0.6380
		9	0.0614	0.0021	0.0073	0.1768	0.5954	0.0076	0.0001	0.0008	0.0001	0.6380	0.0614	0.0021	0.0073	0.5954	0.0076	0.0001	0.0008	0.1658	0.6380
573	8	7	0.897	6.26	89.99	0.1274	6.13	-0.07	0.16	-0.03	0.9992	4.2303	0.0904	0.0020	0.0107	0.5949	-0.0000	0.06	0.0000	1.3644	1.5524
		9	0.0619	0.0020	0.0075	0.1674	0.6065	-0.0013	0.0003	0.0004	0.9992	4.2303	0.0619	0.0020	0.0107	0.5949	-0.0000	0.06	0.0000	1.3644	1.5524
574	1	7	0.896	9.36	11.20	0.0542	6.07	0.0916	0.09	0.06	0.1247	0.6157	0.0617	0.0006	0.0082	0.6672	0.0840	0.0022	0.0098	0.1312	0.5882
		9	0.0185	0.0009	0.0023	0.2670	0.6230	0.0840	0.0022	0.0098	0.1312	0.6157	0.0185	0.0009	0.0023	0.6230	0.0840	0.0022	0.0098	0.1312	0.5882
574	2	7	0.898	9.36	22.50	0.0621	6.10	0.0894	0.12	0.06	0.1115	0.6334	0.0790	0.0009	0.0103	0.6553	0.0787	0.0019	0.0113	0.1441	0.5881
		9	0.0367	0.0017	0.0045	0.2396	0.6185	0.0787	0.0022	0.0092	0.1441	0.6334	0.0367	0.0017	0.0045	0.6185	0.0787	0.0022	0.0092	0.1441	0.5881
574	3	7	0.898	9.36	33.70	0.0728	6.12	0.0805	0.15	0.06	0.1036	0.6449	0.0937	0.0013	0.0120	0.6417	0.0805	0.0016	0.0103	0.1669	0.6006
		9	0.0509	0.0023	0.0060	0.2315	0.5947	0.0704	0.0023	0.0084	0.1669	0.6449	0.0509	0.0023	0.0060	0.5947	0.0704	0.0023	0.0084	0.1669	0.6006

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

See Key →

Run Pt.	Cd	0.897 0.1013 0.0617	9.35 0.0017 0.0027	45.00 0.0123 0.0075	0.0882 0.2216	6.14 0.6085 0.6072	0.01 0.0668 0.0594	0.17 0.0013 0.0021	0.04 0.0087 0.0072	0.1023 0.1843	0.6500 0.6081
574	4 5 9	0.898 0.1074 0.0720	9.36 0.0020 0.0028	56.19 0.0128 0.0084	0.0952 0.1947	6.14 0.5970 0.5882	-0.00 0.0490 0.0467	0.18 0.0011 0.0017	0.02 0.0064 0.0059	0.1153 0.1891	0.6543 0.6388
574	6 7 9	0.897 0.1109 0.0783	9.36 0.0021 0.0028	67.49 0.0130 0.0093	0.0990 0.1835	6.15 0.5892 0.5974	-0.02 0.0305 0.0299	0.19 0.0008 0.0011	0.01 0.0038 0.0037	0.1342 0.1911	0.6355 0.6311
574	7 8 9	0.897 0.1106 0.0841	9.37 0.0023 0.0027	78.69 0.0129 0.0098	0.1072 0.1642	6.15 0.5846 0.5842	-0.05 0.0134 0.0130	0.19 0.0004 0.0004	-0.01 0.0016 0.0015	0.1507 0.1584	0.6164 0.6072
574	8 9	0.900 0.1079 0.0868	9.37 0.0024 0.0026	89.99 0.0125 0.0102	0.1112 0.1506	6.15 0.5818 0.5884	-0.07 0.0009 -0.0011	0.19 -0.0001 -0.0003	-0.04 -0.0001 -0.0004	0.6488 1.4727	0.9593 2.1572
575	1 8 9	0.896 0.0658 0.0228	12.49 0.0006 0.0012	11.20 0.0088 0.0028	0.0510 0.2723	6.08 0.6699 0.6247	0.07 0.1133 0.1037	0.09 0.0024 0.0021	0.07 0.0136 0.0118	0.1086 0.1057	0.6014 0.5707
575	2 8 9	0.900 0.0888 0.0444	12.50 0.0008 0.0022	22.50 0.0116 0.0053	0.0471 0.2577	6.11 0.6557 0.6022	0.07 0.1102 0.0961	0.14 0.0022 0.0023	0.07 0.0134 0.0109	0.1043 0.1204	0.6104 0.5685
575	3 8 9	0.899 0.1087 0.0629	12.49 0.0012 0.0028	33.70 0.0136 0.0072	0.0591 0.2292	6.14 0.6277 0.5772	0.06 0.1025 0.0874	0.17 0.0020 0.0024	0.07 0.0128 0.0099	0.0989 0.1398	0.6273 0.5712
575	4 8 9	0.897 0.1184 0.0769	12.48 0.0017 0.0030	45.00 0.0142 0.0089	0.0756 0.1990	6.15 0.6009 0.5832	0.03 0.0880 0.0762	0.19 0.0015 0.0025	0.06 0.0113 0.0088	0.0872 0.1658	0.6469 0.5794
575	5 8 9	0.897 0.1256 0.0865	12.49 0.0019 0.0030	56.19 0.0147 0.0101	0.0792 0.1739	6.16 0.5874 0.5843	0.01 0.0650 0.0608	0.20 0.0012 0.0023	0.05 0.0085 0.0072	0.0944 0.1942	0.6543 0.5943
575	6 8 9	0.898 0.1266 0.0963	12.48 0.0020 0.0028	67.48 0.0147 0.0111	0.0795 0.1482	6.16 0.5810 0.5799	-0.01 0.0407 0.0427	0.21 0.0009 0.0017	0.03 0.0052 0.0054	0.1105 0.2016	0.6465 0.6353

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key											
575	7 8 9	0.899 0.1243 0.1069	12.49 0.0020 0.0026	78.69 0.0145 0.0119	0.0819 0.1226	6.16 0.5862 0.5576	-0.04 0.0172 0.0195	0.21 0.0004 0.0007	-0.00 0.0021 0.0024	0.1415 0.1862	0.6152 0.6256		
575	7 8 9	0.898 0.1210 0.1090	12.49 0.0018 0.0027	90.00 0.0139 0.0127	0.0764 0.1245	6.15 0.5758 0.5833	-0.07 -0.0017 -0.0009	0.22 -0.0001 -0.0002	-0.02 -0.0003 -0.0003	0.4252 1.5956	0.9071 1.8435		
576	7 8 9	0.897 0.0536 -0.0043	-3.09 0.0016 -0.0000	11.19 0.0070 -0.0005	0.1512 0.0172	9.08 0.6602 0.6716	-0.10 -0.0258 -0.0301	0.05 -0.0008 -0.0008	-0.06 -0.0035 -0.0041	0.1566 0.1411	0.6863 0.6916		
576	7 8 9	0.898 0.0485 -0.0098	-3.07 0.0013 -0.0000	22.49 0.0064 -0.0013	0.1421 0.0342	9.07 0.6688 0.7110	-0.10 -0.0240 -0.0284	0.04 -0.0007 -0.0007	-0.05 -0.0032 -0.0039	0.1622 0.1369	0.6777 0.6962		
576	7 8 9	0.898 0.0430 -0.0153	-3.07 0.0011 -0.0001	33.69 0.0058 -0.0020	0.1336 0.0558	9.06 0.6838 0.6774	-0.10 -0.0207 -0.0254	0.03 -0.0007 -0.0006	-0.05 -0.0028 -0.0035	0.1719 0.1324	0.6941 0.6982		
576	7 8 9	0.899 0.0388 -0.0206	-3.07 0.0008 -0.0002	44.99 0.0053 -0.0026	0.1086 0.0667	9.06 0.6830 0.6477	-0.09 -0.0173 -0.0213	0.03 -0.0005 -0.0005	-0.04 -0.0023 -0.0029	0.1717 0.1256	0.6909 0.6965		
576	7 8 9	0.898 0.0342 -0.0242	-3.07 0.0007 -0.0004	56.19 0.0049 -0.0033	0.1149 0.0915	9.05 0.7261 0.6886	-0.09 -0.0124 -0.0157	0.02 -0.0005 -0.0003	-0.04 -0.0018 -0.0022	0.2084 0.1259	0.7297 0.7207		
576	7 8 9	0.898 0.0320 -0.0287	-3.06 0.0005 -0.0004	67.49 0.0045 -0.0036	0.0903 0.0752	9.05 0.7155 0.6351	-0.08 -0.0073 -0.0102	0.02 -0.0004 -0.0002	-0.04 -0.0011 -0.0015	0.2855 0.1458	0.7974 0.7599		
576	7 8 9	0.900 0.0306 -0.0300	-3.05 0.0004 -0.0005	78.69 0.0042 -0.0040	0.0713 0.0942	9.04 0.6477 0.6716	-0.07 -0.0018 -0.0045	0.01 -0.0002 -0.0002	-0.02 -0.0004 -0.0008	0.7977 0.2677	1.2966 0.9578		
576	7 8 9	0.897 0.0299 -0.0321	-3.06 0.0003 -0.0005	89.99 0.0042 -0.0041	0.0653 0.0874	9.04 0.7036 0.6403	-0.06 0.0032 0.0007	0.00 -0.0001 -0.0001	-0.03 0.0003 -0.0001	-0.1548 -1.1822	0.5629 -0.6411		
577	7 8 9	0.898 0.0628 0.0014	0.02 0.0014 0.0001	11.19 0.0080 0.0000	0.1134 0.6648	9.09 0.6370 0.0323	-0.06 0.0042 0.0009	0.06 0.0001 -0.0001	-0.02 0.0006 0.0000	0.1555 -1.0245	0.7640 0.3292		

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																			
577	2	7	0.898	0.02	22.49	0.1187	0.6459	-0.06	0.06	-0.003	0.1334	0.7420	0.0617	0.0014	0.0079	0.0040	0.0001	0.0005	0.0000	0.0000	-1.0122	0.2351
	6	9	0.0013	0.0002	0.0000	0.8056	0.3472	0.0009	-0.0001	0.0000	-0.0000	0.0000	0.7420	0.0013	0.0002	0.0000	0.0009	-0.0001	0.0000	0.0000	-1.0122	0.2351
	9	7	0.897	0.03	33.69	0.1159	0.6443	-0.06	0.06	-0.003	0.0569	0.6904	0.0616	0.0014	0.0079	0.0035	0.0000	0.0004	0.0000	0.0000	0.0569	0.6904
577	3	7	0.897	0.03	33.69	0.1159	0.6443	-0.06	0.06	-0.003	0.0569	0.6904	0.0616	0.0014	0.0079	0.0035	0.0000	0.0004	0.0000	0.0000	0.0569	0.6904
	6	9	0.0005	0.0002	0.0002	2.8761	2.2527	0.0007	-0.0001	0.0000	-0.0000	0.2542	0.6904	0.0005	0.0002	0.0007	0.0000	0.0000	0.0000	0.0000	-1.2000	0.2542
	9	7	0.898	0.03	44.99	0.1130	0.6456	-0.07	0.06	-0.003	0.0408	0.6788	0.0607	0.0013	0.0078	0.0007	0.0000	0.0000	0.0000	0.0000	-0.0408	0.6788
577	4	7	0.898	0.03	44.99	0.1130	0.6456	-0.07	0.06	-0.003	0.0408	0.6788	0.0607	0.0013	0.0078	0.0007	0.0000	0.0000	0.0000	0.0000	-0.0408	0.6788
	6	9	0.0007	0.0002	0.0001	1.9330	1.1497	0.0007	-0.0002	0.0000	-0.0000	0.1186	0.6788	0.0007	0.0001	0.0007	0.0000	0.0000	0.0000	0.0000	-1.4656	0.1186
	9	7	0.897	0.03	56.19	0.1185	0.6573	-0.07	0.06	-0.003	0.2041	0.5929	0.0593	0.0014	0.0078	0.0021	0.0000	0.0002	0.0000	0.0000	0.2041	0.5929
577	5	7	0.897	0.03	56.19	0.1185	0.6573	-0.07	0.06	-0.003	0.2041	0.5929	0.0593	0.0014	0.0078	0.0021	0.0000	0.0002	0.0000	0.0000	0.2041	0.5929
	6	9	0.0009	0.0002	0.0000	1.2547	0.4117	0.0007	-0.0002	0.0000	-0.0000	0.1492	0.5929	0.0009	0.0002	0.0007	0.0000	0.0000	0.0000	0.0000	-1.4167	0.1492
	9	7	0.898	0.03	67.49	0.1139	0.6576	-0.06	0.05	-0.003	0.5255	0.4395	0.0590	0.0013	0.0077	0.0015	0.0001	0.0000	0.0000	0.0000	0.5255	0.4395
577	6	7	0.898	0.03	67.49	0.1139	0.6576	-0.06	0.05	-0.003	0.5255	0.4395	0.0590	0.0013	0.0077	0.0015	0.0001	0.0000	0.0000	0.0000	0.5255	0.4395
	6	9	0.0590	0.0013	0.0077	2.3764	1.3431	0.0007	-0.0002	0.0000	-0.0000	0.2542	0.4395	0.0590	0.0013	0.0077	0.0015	0.0001	0.0000	0.0000	-1.7662	0.2542
	9	7	0.898	0.04	78.69	0.1148	0.6595	-0.07	0.06	-0.004	0.0162	0.1900	0.0587	0.0013	0.0077	0.0009	0.0001	0.0000	0.0000	0.0000	0.0162	0.1900
577	7	7	0.898	0.04	78.69	0.1148	0.6595	-0.07	0.06	-0.004	0.0162	0.1900	0.0587	0.0013	0.0077	0.0009	0.0001	0.0000	0.0000	0.0000	0.0162	0.1900
	6	9	0.0002	0.0002	0.0001	4.7187	2.8686	0.0004	-0.0002	0.0000	-0.0000	0.0124	0.1900	0.0002	0.0002	0.0004	0.0001	0.0000	0.0000	0.0000	-3.0094	0.0124
	9	7	0.898	0.04	89.99	0.1126	0.6525	-0.06	0.06	-0.003	0.4788	0.2583	0.0590	0.0013	0.0077	0.0007	0.0000	0.0000	0.0000	0.0000	0.4788	0.2583
577	8	7	0.898	0.04	89.99	0.1126	0.6525	-0.06	0.06	-0.003	0.4788	0.2583	0.0590	0.0013	0.0077	0.0007	0.0000	0.0000	0.0000	0.0000	0.4788	0.2583
	6	9	0.0590	0.0013	0.0077	2.1251	1.1281	0.0001	-0.0002	0.0000	-0.0002	0.0694	0.2583	0.0590	0.0013	0.0077	0.0001	0.0000	0.0000	0.0000	-13.0515	0.0694
	9	7	0.896	1.05	11.20	0.1130	0.6360	-0.04	0.06	-0.001	0.1223	0.6954	0.0652	0.0014	0.0082	0.0139	0.0002	0.0010	0.0010	0.0010	0.1223	0.6954
578	1	7	0.896	1.05	11.20	0.1130	0.6360	-0.04	0.06	-0.001	0.1223	0.6954	0.0652	0.0014	0.0082	0.0139	0.0002	0.0010	0.0010	0.0010	0.1223	0.6954
	6	9	0.0026	0.0003	0.0003	0.6065	0.6679	0.0077	0.0002	0.0000	0.0000	0.6954	0.6954	0.0026	0.0003	0.0077	0.0002	0.0000	0.0000	0.0000	0.1223	0.6954
	9	7	0.896	1.05	22.50	0.1114	0.6302	-0.04	0.07	-0.001	0.1370	0.7275	0.0665	0.0014	0.0083	0.0122	0.0002	0.0010	0.0010	0.0010	0.1370	0.7275
578	2	7	0.896	1.05	22.50	0.1114	0.6302	-0.04	0.07	-0.001	0.1370	0.7275	0.0665	0.0014	0.0083	0.0122	0.0002	0.0010	0.0010	0.0010	0.1370	0.7275
	6	9	0.0053	0.0003	0.0007	0.3244	0.6707	0.0074	0.0002	0.0000	0.0000	0.6954	0.6954	0.0053	0.0003	0.0074	0.0002	0.0000	0.0000	0.0000	0.1370	0.7275
	9	7	0.896	1.06	33.69	0.1190	0.6395	-0.05	0.06	-0.001	0.1451	0.7668	0.0667	0.0015	0.0085	0.0100	0.0002	0.0008	0.0008	0.0008	0.1451	0.7668
578	3	7	0.896	1.06	33.69	0.1190	0.6395	-0.05	0.06	-0.001	0.1451	0.7668	0.0667	0.0015	0.0085	0.0100	0.0002	0.0008	0.0008	0.0008	0.1451	0.7668
	6	9	0.0073	0.0003	0.0010	0.2672	0.6826	0.0064	0.0001	0.0000	0.0000	0.6954	0.6954	0.0073	0.0003	0.0064	0.0001	0.0000	0.0000	0.0000	0.1451	0.7668
	9	7	0.898	1.06	44.99	0.1181	0.6303	-0.05	0.07	-0.001	0.1899	0.7174	0.0670	0.0015	0.0084	0.0072	0.0002	0.0010	0.0010	0.0010	0.1899	0.7174
578	4	7	0.898	1.06	44.99	0.1181	0.6303	-0.05	0.07	-0.001	0.1899	0.7174	0.0670	0.0015	0.0084	0.0072	0.0002	0.0010	0.0010	0.0010	0.1899	0.7174
	6	9	0.0085	0.0004	0.0012	0.2696	0.7151	0.0054	0.0001	0.0000	0.0000	0.6954	0.6954	0.0085	0.0004	0.0054	0.0001	0.0000	0.0000	0.0000	0.1899	0.7174
	9	7	0.898	1.06	56.19	0.1181	0.6303	-0.05	0.07	-0.001	0.1899	0.7174	0.0670	0.0015	0.0084	0.0072	0.0002	0.0010	0.0010	0.0010	0.1899	0.7174

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																						
578	5	7	0.849	1.06	56.19	0.1171	0.6294	9.09	-0.06	0.07	-0.01	0.1640	0.7107	0.0672	0.0015	0.0084	0.0013	0.2442	0.6977	0.0050	0.0001	0.0004	0.0467	0.5942
578	6	7	0.896	1.07	67.49	0.1202	0.6316	9.09	-0.07	0.07	-0.02	-0.0066	0.5968	0.0670	0.0016	0.0084	0.0015	0.2500	0.7209	0.0035	0.0000	0.0004	-0.1270	0.4401
578	7	7	0.895	1.07	78.69	0.1215	0.6326	9.09	-0.07	0.07	-0.02	-0.3752	0.3693	0.0670	0.0016	0.0084	0.0015	0.2273	0.6893	0.0019	0.0002	0.0000	-0.5371	0.2203
578	8	7	0.899	1.07	89.99	0.1265	0.6353	9.10	-0.07	0.07	-0.02	-6.8540	-3.2941	0.0663	0.0016	0.0084	0.0016	0.2394	0.7323	0.0001	0.0002	-0.0001	-6.8540	-3.2941
579	1	7	0.896	3.12	11.20	0.1020	0.6391	9.09	-0.01	0.07	0.00	0.1295	0.6740	0.0686	0.0014	0.0087	0.0008	0.3096	0.5692	0.0346	0.0006	0.0037	0.1234	0.6568
579	2	7	0.898	3.12	22.50	0.1080	0.6318	9.10	-0.02	0.08	0.00	0.1305	0.6669	0.0125	0.0007	0.0016	0.0016	0.2901	0.6731	0.0261	0.0006	0.0034	0.1273	0.6516
579	3	7	0.895	3.13	33.70	0.1060	0.6184	9.10	-0.02	0.09	-0.00	0.1370	0.6693	0.0773	0.0016	0.0095	0.0025	0.2502	0.6919	0.0254	0.0006	0.0030	0.1343	0.6574
579	4	7	0.897	3.13	45.00	0.1154	0.6154	9.11	-0.03	0.09	-0.00	0.1296	0.6633	0.0245	0.0018	0.0031	0.0031	0.2078	0.6440	0.0188	0.0005	0.0024	0.1336	0.6572
579	5	7	0.901	3.13	56.20	0.1166	0.6147	9.11	-0.04	0.10	-0.01	0.1426	0.6489	0.0813	0.0018	0.0100	0.0038	0.1927	0.6692	0.0139	0.0003	0.0018	0.1053	0.6507
579	6	7	0.902	3.13	67.49	0.1219	0.6114	9.12	-0.05	0.11	-0.02	0.1414	0.6736	0.0326	0.0011	0.0042	0.0042	0.1708	0.6462	0.0083	0.0002	0.0010	0.1195	0.6224
579	7	7	0.902	3.13	78.69	0.1265	0.6111	9.12	-0.06	0.10	-0.02	0.0268	0.6179	0.0344	0.0010	0.0043	0.0043	0.1588	0.6344	0.0042	0.0000	0.0003	0.0268	0.4450

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key														
579	6	0.9012	0.0020	0.0011	0.1284	0.6061	0.6581	-0.0001	-0.0006	-0.0001	0.112	-0.0003	-0.0003	-0.0001	7.1081	3.3401
	8	0.0812	0.0020	0.0011	0.1284	0.6061	0.6581	-0.0001	-0.0006	-0.0001	0.112	-0.0003	-0.0003	-0.0001	2.6021	2.2836
	9	0.0333	0.0011	0.0007	0.1687	0.6581	0.6581	-0.0006	-0.0006	-0.0003	0.0003	-0.0003	-0.0003	-0.0003		
580	1	0.897	0.0010	0.0007	0.0730	0.6465	0.5919	0.0659	0.0609	0.0015	0.0017	0.0017	0.0017	0.04	0.1210	0.6518
	6	0.0749	0.0010	0.0007	0.0730	0.6465	0.5919	0.0659	0.0609	0.0015	0.0017	0.0017	0.0017	0.0086	0.1441	0.6313
	9	0.0137	0.0007	0.0007	0.2873	0.5919	0.5919	0.0609	0.0609	0.0017	0.0017	0.0017	0.0017	0.0076		
580	2	0.899	0.0013	0.0013	0.0798	0.6297	0.6694	0.0611	0.0558	0.0013	0.0017	0.0017	0.0017	0.03	0.1133	0.6616
	6	0.0851	0.0013	0.0013	0.0798	0.6297	0.6694	0.0611	0.0558	0.0013	0.0017	0.0017	0.0017	0.0080	0.1539	0.6331
	9	0.0250	0.0013	0.0013	0.2766	0.6694	0.6694	0.0558	0.0558	0.0017	0.0017	0.0017	0.0017	0.0071		
580	3	0.898	0.0016	0.0017	0.0901	0.6112	0.6627	0.0634	0.0491	0.0012	0.0016	0.0016	0.0016	0.03	0.1124	0.6638
	6	0.0915	0.0016	0.0017	0.0901	0.6112	0.6627	0.0634	0.0491	0.0012	0.0016	0.0016	0.0016	0.0070	0.1651	0.6470
	9	0.0374	0.0017	0.0017	0.2356	0.6627	0.6627	0.0491	0.0491	0.0016	0.0016	0.0016	0.0016	0.0063		
580	4	0.900	0.0019	0.0020	0.1014	0.6050	0.6257	0.0626	0.0388	0.0010	0.0013	0.0013	0.0013	0.02	0.1211	0.6591
	6	0.0970	0.0019	0.0020	0.1014	0.6050	0.6257	0.0626	0.0388	0.0010	0.0013	0.0013	0.0013	0.0056	0.1774	0.6469
	9	0.0469	0.0020	0.0023	0.2190	0.6257	0.6257	0.0388	0.0388	0.0013	0.0013	0.0013	0.0013	0.0050		
580	5	0.900	0.0021	0.0023	0.1064	0.5982	0.6352	0.0612	0.0283	0.0015	0.0016	0.0016	0.0016	0.00	0.1271	0.6417
	6	0.1003	0.0021	0.0023	0.1064	0.5982	0.6352	0.0612	0.0283	0.0015	0.0016	0.0016	0.0016	0.0040	0.1807	0.6404
	9	0.0518	0.0023	0.0023	0.2256	0.6352	0.6352	0.0283	0.0283	0.0016	0.0016	0.0016	0.0016	0.0036		
580	6	0.897	0.0022	0.0022	0.1074	0.5891	0.6065	0.06198	0.0175	0.0015	0.0015	0.0015	0.0015	0.00	0.1349	0.6265
	6	0.1026	0.0022	0.0022	0.1074	0.5891	0.6065	0.06198	0.0175	0.0015	0.0015	0.0015	0.0015	0.0024	0.1619	0.6474
	9	0.0583	0.0022	0.0022	0.1970	0.6065	0.6065	0.0175	0.0175	0.0015	0.0015	0.0015	0.0015	0.0022		
580	7	0.898	0.0023	0.0022	0.1126	0.5857	0.6188	0.06089	0.0067	0.0016	0.0016	0.0016	0.0016	0.03	0.1311	0.6058
	6	0.1025	0.0023	0.0022	0.1126	0.5857	0.6188	0.06089	0.0067	0.0016	0.0016	0.0016	0.0016	0.0010	0.0738	0.5557
	9	0.0603	0.0022	0.0022	0.1889	0.6188	0.6188	0.0067	0.0067	0.0016	0.0016	0.0016	0.0016	0.0007		
580	8	0.899	0.0024	0.0021	0.1205	0.5832	0.6181	0.0606	0.0025	0.0016	0.0016	0.0016	0.0016	0.03	5.8552	3.9668
	6	0.1009	0.0024	0.0021	0.1205	0.5832	0.6181	0.0606	0.0025	0.0016	0.0016	0.0016	0.0016	0.0001	0.7109	1.0662
	9	0.0613	0.0021	0.0021	0.1750	0.6181	0.6181	0.0025	0.0025	0.0016	0.0016	0.0016	0.0016	0.0005		
581	1	0.900	0.0008	0.0011	0.0510	0.6498	0.7088	0.0635	0.0831	0.0021	0.0022	0.0022	0.0022	0.06	0.1164	0.6212
	6	0.0820	0.0008	0.0011	0.0510	0.6498	0.7088	0.0635	0.0831	0.0021	0.0022	0.0022	0.0022	0.0116	0.1344	0.5873
	9	0.0175	0.0011	0.0011	0.3265	0.7088	0.7088	0.0831	0.0831	0.0022	0.0022	0.0022	0.0022	0.0097		
581	2	0.899	0.0012	0.0018	0.0619	0.6404	0.6283	0.064	0.0774	0.0012	0.0018	0.0018	0.0018	0.06	0.1051	0.6352
	6	0.0975	0.0012	0.0018	0.0619	0.6404	0.6283	0.064	0.0774	0.0018	0.0018	0.0018	0.0018	0.0114	0.1494	0.5869
	9	0.0366	0.0018	0.0018	0.2498	0.6283	0.6283	0.0774	0.0774	0.0018	0.0018	0.0018	0.0018	0.0090		

TABLE VI
 CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd																				
581	3	7 6 9	0.0999	0.0016	9.35	33.70	0.0766	0.2475	9.13	0.6068	0.0808	0.0015	0.0023	0.0063	0.0104	0.0976	0.1695	0.6452	0.5978			
			0.0499	0.0024	0.0024	0.0061	0.0061	0.00694	0.6172	0.5954	0.0664	0.0021	0.0071	0.0083	0.0687							
581	4	7 6 9	0.098	0.0019	9.35	45.00	0.0836	0.2282	9.14	0.5954	0.0664	0.0021	0.0071	0.0083	0.0687	0.0985	0.1879	0.6578	0.6105			
			0.0614	0.0028	0.0028	0.0075	0.0075	0.0585	0.6159	0.5864	0.0483	0.0017	0.0058	0.0071	0.0687							
581	5	7 6 9	0.098	0.0021	9.35	56.19	0.0892	0.2006	9.15	0.5864	0.0483	0.0017	0.0058	0.0071	0.0687	0.1126	0.1929	0.6524	0.6374			
			0.0716	0.0028	0.0028	0.0085	0.0085	0.0455	0.5950	0.5864	0.0455	0.0017	0.0058	0.0071	0.0687							
581	6	7 6 9	0.098	0.0021	9.35	67.49	0.0891	0.1824	9.15	0.5773	0.0302	0.0007	0.0011	0.0037	0.0036	0.1246	0.1963	0.6278	0.6366			
			0.0787	0.0028	0.0028	0.0093	0.0093	0.0290	0.5917	0.5773	0.0290	0.0007	0.0011	0.0037	0.0036							
581	7	7 6 9	0.099	0.0020	9.36	78.69	0.0870	0.1648	9.15	0.5775	0.0129	0.0003	0.0003	0.0015	0.0014	0.1339	0.1608	0.6046	0.6121			
			0.0843	0.0027	0.0027	0.0099	0.0099	0.0116	0.5872	0.5775	0.0116	0.0003	0.0003	0.0015	0.0014							
581	8	7 6 9	0.900	0.0020	9.36	89.99	0.0890	0.1544	9.14	0.5790	0.0016	0.0001	0.0003	0.0006	0.0006	0.5375	0.8360	1.0573	1.4699			
			0.1355	0.0026	0.0026	0.0103	0.0103	0.0021	0.5995	0.5790	0.0016	0.0001	0.0003	0.0006	0.0006							
582	1	7 6 9	0.098	0.0007	12.48	11.20	0.0428	0.2863	9.09	0.6586	0.1146	0.0023	0.0022	0.0016	0.0016	0.1035	0.1085	0.6044	0.5701			
			0.0228	0.0013	0.0013	0.0029	0.0029	0.1018	0.6345	0.6345	0.1018	0.0022	0.0022	0.0016	0.0016							
582	2	7 6 9	0.901	0.0009	12.47	22.50	0.0425	0.2638	9.12	0.6470	0.1110	0.0023	0.0023	0.0017	0.0017	0.1018	0.1223	0.6122	0.5665			
			0.1079	0.0023	0.0023	0.0053	0.0053	0.0948	0.6010	0.6470	0.1110	0.0023	0.0023	0.0017	0.0017							
582	3	7 6 9	0.097	0.0017	12.47	33.71	0.0726	0.2379	9.14	0.6023	0.1027	0.0024	0.0024	0.0019	0.0019	0.0950	0.1428	0.6284	0.5730			
			0.175	0.0029	0.0029	0.0073	0.0073	0.0865	0.5870	0.6023	0.1027	0.0024	0.0024	0.0019	0.0019							
582	4	7 6 9	0.099	0.0019	12.48	45.00	0.0771	0.1983	9.15	0.5952	0.0669	0.0025	0.0025	0.0019	0.0019	0.0841	0.1698	0.6470	0.5798			
			0.1273	0.0030	0.0030	0.0089	0.0089	0.0746	0.5768	0.5952	0.0669	0.0025	0.0025	0.0019	0.0019							
582	5	7 6 9	0.098	0.0018	12.46	56.20	0.0693	0.1728	9.16	0.5765	0.0637	0.0023	0.0023	0.0019	0.0019	0.0935	0.2004	0.6511	0.6015			
			0.1321	0.0030	0.0030	0.0100	0.0100	0.0592	0.5771	0.5765	0.0637	0.0023	0.0023	0.0019	0.0019							

See Key

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																				
582	6	0.901	12.46	67.48	0.0837	0.5636	-0.091	0.0008	0.0049	0.1089	0.6357	0.1254	0.0021	0.0148	0.0108	0.1388	0.5523	0.0411	0.0016	0.0051	0.1994	0.6231
	6	0.0985	0.0027	0.0108	0.1388	0.5523	0.0411	0.0016	0.0051	0.1994	0.6231	0.0985	0.0027	0.0108	0.1388	0.5523	0.0411	0.0016	0.0051	0.1994	0.6231	
582	7	0.902	12.47	78.69	0.0984	0.6018	-0.04	0.0004	-0.000	0.1433	0.6036	0.1191	0.0023	0.0143	0.0116	0.1210	0.5452	0.0180	0.0006	0.0022	0.1812	0.6139
	6	0.1071	0.0025	0.0116	0.1210	0.5452	0.0180	0.0006	0.0022	0.1812	0.6139	0.1071	0.0025	0.0116	0.1210	0.5452	0.0180	0.0006	0.0022	0.1812	0.6139	
582	8	0.902	12.47	90.00	0.0961	0.5909	-0.07	0.0002	-0.0004	0.4967	1.0270	0.1163	0.0022	0.0124	0.0124	0.0961	0.5909	0.0017	0.0003	0.0005	1.0116	1.4532
	6	0.1107	0.0025	0.0124	0.1170	0.5608	0.0017	0.0003	0.0005	1.0116	1.4532	0.1107	0.0025	0.0124	0.0124	0.0961	0.5909	0.0017	0.0003	0.0005	1.0116	1.4532
583	1	0.901	-3.08	11.19	0.1563	0.5967	-0.11	0.0008	-0.0006	0.1658	0.6791	0.0806	0.0025	0.0096	0.0007	0.1563	0.5967	0.0255	0.0008	0.0043	0.1339	0.6756
	8	-0.0030	-0.0000	-0.0007	0.1563	0.5967	0.0255	0.0008	0.0043	0.1339	0.6756	-0.0030	-0.0000	-0.0007	0.0007	0.1563	0.5967	0.0255	0.0008	0.0043	0.1339	0.6756
583	2	0.900	-3.07	22.49	0.1553	0.6037	-0.10	0.0008	-0.0032	0.1749	0.6830	0.0773	0.0024	0.0093	0.0013	0.1553	0.6037	0.0303	0.0007	0.0041	0.1306	0.6759
	6	-0.0096	-0.0000	-0.0013	0.1553	0.6037	0.0303	0.0007	0.0041	0.1306	0.6759	-0.0096	-0.0000	-0.0013	0.0013	0.1553	0.6037	0.0303	0.0007	0.0041	0.1306	0.6759
583	3	0.899	-3.05	33.69	0.1497	0.6095	-0.09	0.0004	-0.0037	0.1764	0.6852	0.0740	0.0022	0.0090	0.0020	0.1497	0.6095	0.0274	0.0007	0.0037	0.1293	0.6822
	6	-0.0148	-0.0001	-0.0020	0.1497	0.6095	0.0274	0.0007	0.0037	0.1293	0.6822	-0.0148	-0.0001	-0.0020	0.0020	0.1497	0.6095	0.0274	0.0007	0.0037	0.1293	0.6822
583	4	0.899	-3.06	44.99	0.1465	0.6207	-0.09	0.0006	-0.0031	0.1913	0.7053	0.0710	0.0020	0.0088	0.0026	0.1465	0.6207	0.0167	0.0006	0.0031	0.1300	0.7053
	6	-0.0196	-0.0002	-0.0026	0.1465	0.6207	0.0167	0.0006	0.0031	0.1300	0.7053	-0.0196	-0.0002	-0.0026	0.0026	0.1465	0.6207	0.0167	0.0006	0.0031	0.1300	0.7053
583	5	0.900	-3.05	56.19	0.1376	0.6284	-0.08	0.0005	-0.0004	0.2431	0.7469	0.0687	0.0018	0.0086	0.0032	0.1376	0.6284	0.0177	0.0004	0.0025	0.1252	0.7469
	6	-0.0234	-0.0004	-0.0032	0.1376	0.6284	0.0177	0.0004	0.0025	0.1252	0.7469	-0.0234	-0.0004	-0.0032	0.0032	0.1376	0.6284	0.0177	0.0004	0.0025	0.1252	0.7469
583	6	0.900	-3.03	67.49	0.1297	0.6380	-0.07	0.0004	-0.0017	0.3450	0.8114	0.0676	0.0017	0.0086	0.0036	0.1297	0.6380	0.0121	0.0003	0.0017	0.1324	0.8114
	6	-0.0272	-0.0004	-0.0036	0.1297	0.6380	0.0121	0.0003	0.0017	0.1324	0.8114	-0.0272	-0.0004	-0.0036	0.0036	0.1297	0.6380	0.0121	0.0003	0.0017	0.1324	0.8114
583	7	0.899	-3.04	78.69	0.1244	0.6439	-0.07	0.0003	-0.0004	1.3883	1.5812	0.0669	0.0016	0.0086	0.0040	0.1244	0.6439	0.0013	0.0003	0.0004	1.1842	1.5812
	6	-0.0288	-0.0005	-0.0040	0.1244	0.6439	0.0013	0.0003	0.0004	1.1842	1.5812	-0.0288	-0.0005	-0.0040	0.0040	0.1244	0.6439	0.0013	0.0003	0.0004	1.1842	1.5812
583	8	0.901	-3.04	89.99	0.1156	0.6434	-0.06	0.0001	-0.0002	-0.2111	2.6077	0.0666	0.0015	0.0085	0.0040	0.1156	0.6434	0.0034	0.0002	0.0003	-0.2111	2.6077
	6	-0.0305	-0.0005	-0.0040	0.1156	0.6434	0.0034	0.0002	0.0003	-0.2111	2.6077	-0.0305	-0.0005	-0.0040	0.0040	0.1156	0.6434	0.0034	0.0002	0.0003	-0.2111	2.6077

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key											
584	1	7	0.899	0.022	11.19	0.0105	0.1257	15.14	-0.06	0.06	-0.03	0.0813	0.7110
		8	0.0893	0.0022	0.0001	0.0001	0.5923	0.0043	0.0000	0.0000	0.0006	2.1067	1.3171
		9	0.0010	0.0002	0.0001	0.0001	0.5175	-0.0006	-0.0002	-0.0001	-0.0001		
584	2	7	0.896	0.023	22.49	0.0106	0.1295	15.14	-0.06	0.06	-0.03	0.0485	0.6818
		8	0.0888	0.0023	0.0001	0.0001	0.5987	0.0040	0.0000	0.0005	0.0005	2.4204	1.7386
		9	0.0016	0.0002	0.0001	0.0001	0.4501	-0.0005	-0.0002	-0.0001	-0.0001		
584	3	7	0.899	0.04	33.69	0.0105	0.1324	15.15	-0.05	0.06	-0.03	-0.0061	0.6721
		8	0.0878	0.0023	0.0001	0.0001	0.6210	0.0035	-0.0002	-0.0004	0.0004	2.1601	1.5169
		9	0.0016	0.0002	0.0001	0.0001	0.5407	-0.0006	-0.0002	-0.0001	-0.0001		
584	4	7	0.899	0.04	44.99	0.0104	0.1310	15.14	-0.07	0.06	-0.03	-0.1357	0.5836
		8	0.0871	0.0022	0.0001	0.0001	0.6705	0.0028	-0.0002	-0.0002	0.0003	2.6190	1.9673
		9	0.0018	0.0002	0.0001	0.0001	0.3475	-0.0005	-0.0002	-0.0002	-0.0002		
584	5	7	0.900	0.05	56.19	0.0103	0.1346	15.14	-0.06	0.06	-0.04	-0.3459	0.4310
		8	0.0862	0.0023	0.0001	0.0001	0.9060	0.0021	-0.0001	0.0001	0.0001	1.8859	1.5307
		9	0.0013	0.0002	0.0001	0.0001	0.5898	-0.0007	-0.0002	-0.0002	-0.0002		
584	6	7	0.900	0.04	67.49	0.0102	0.1390	15.14	-0.06	0.06	-0.03	-0.7210	0.1722
		8	0.0850	0.0023	0.0001	0.0001	0.9787	0.0014	-0.0002	-0.0002	0.0000	1.7507	1.6407
		9	0.0013	0.0002	0.0001	0.0001	0.5343	-0.0008	-0.0003	-0.0003	-0.0002		
584	7	7	0.897	0.04	78.69	0.0102	0.1311	15.14	-0.06	0.06	-0.03	-1.6034	-0.0762
		8	0.0860	0.0022	0.0002	0.0002	1.2612	0.0007	-0.0002	-0.0003	0.0000	1.2136	1.4340
		9	0.0011	0.0002	0.0002	0.0002	0.9885	-0.0012	-0.0003	-0.0003	-0.0003		
584	8	7	0.900	0.05	89.99	0.0102	0.1354	15.13	-0.06	0.06	-0.03	-2.5124	-0.7457
		8	0.0853	0.0023	0.0002	0.0002	1.4104	0.0004	-0.0002	-0.0002	0.0000	0.8073	1.2731
		9	0.0009	0.0002	0.0002	0.0002	1.0456	-0.0018	-0.0003	-0.0004	-0.0004		
585	1	7	0.898	1.05	11.19	0.0109	0.1199	15.14	-0.05	0.06	-0.01	0.1201	0.6851
		8	0.0916	0.0021	0.0003	0.0003	0.4319	0.0133	0.0003	0.0001	0.0018	0.1297	0.6533
		9	0.0035	0.0003	0.0003	0.0003	0.4824	0.0058	0.0001	0.0007	0.0007		
585	2	7	0.899	1.06	22.49	0.0110	0.1209	15.14	-0.05	0.07	-0.01	0.1237	0.6733
		8	0.0925	0.0022	0.0007	0.0007	0.2946	0.0117	0.0002	0.0015	0.0015	0.1529	0.6953
		9	0.0059	0.0003	0.0007	0.0007	0.6164	0.0052	0.0001	0.0007	0.0007		
585	3	7	0.897	1.07	33.69	0.0110	0.1216	15.15	-0.05	0.07	-0.02	0.1276	0.6667
		8	0.0931	0.0022	0.0009	0.0009	0.2550	0.0096	0.0002	0.0012	0.0005	0.1310	0.6413
		9	0.0079	0.0004	0.0009	0.0009	0.6255	0.0046	0.0001	0.0005	0.0005		

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key													
585	4	0.898	1.08	44.99	0.1195	15.15	-0.002	0.07	-0.009	0.1174	0.6476				
	8	0.0935	0.0022	0.0110	0.2705	0.5905	0.0074	0.0001	0.0009	0.0560	0.5841				
	9	0.0089	0.0004	0.0012	0.2705	0.6935	0.0039	0.0000	0.0004	0.0560	0.5841				
585	5	0.898	1.06	56.19	0.1219	15.15	-0.006	0.07	-0.002	0.0752	0.6269				
	8	0.0930	0.0022	0.0110	0.2681	0.5923	0.0050	0.0000	0.0006	-0.0871	0.4659				
	9	0.0098	0.0005	0.0014	0.2681	0.7150	0.0031	-0.0000	0.0002	-0.0871	0.4659				
585	6	0.899	1.08	67.49	0.1252	15.15	-0.006	0.07	-0.002	0.0748	0.5576				
	8	0.0925	0.0023	0.0110	0.2524	0.5945	0.0032	-0.0000	0.0003	-0.0748	0.5576				
	9	0.0109	0.0005	0.0015	0.2524	0.7003	0.0019	-0.0001	0.0001	-0.0748	0.5576				
585	7	0.898	1.08	78.69	0.1246	15.14	-0.006	0.07	-0.003	0.6155	0.2108				
	8	0.0925	0.0023	0.0109	0.2563	0.5934	0.0016	-0.0001	0.0000	-0.6155	0.2108				
	9	0.0112	0.0005	0.0016	0.2563	0.7265	0.0003	-0.0002	-0.0001	-0.6155	0.2108				
585	8	0.897	1.08	89.99	0.1269	15.15	-0.007	0.07	-0.003	17.6660	10.3469				
	8	0.0921	0.0023	0.0109	0.2452	0.5957	-0.0000	-0.0003	-0.0001	17.6660	10.3469				
	9	0.0117	0.0005	0.0016	0.2452	0.6934	-0.0021	-0.0003	-0.0005	17.6660	10.3469				
586	1	0.897	1.13	11.20	0.1060	15.15	-0.001	0.07	-0.000	0.1285	0.6621				
	8	0.0962	0.0020	0.0114	0.3558	0.5933	0.0335	0.0008	0.0044	0.1273	0.6465				
	9	0.0072	0.0005	0.0009	0.3558	0.6691	0.0261	0.0006	0.0033	0.1273	0.6465				
586	2	0.899	1.13	22.50	0.1093	15.15	-0.002	0.08	-0.000	0.1302	0.6542				
	8	0.0922	0.0021	0.0117	0.2651	0.5941	0.0292	0.0007	0.0038	0.1448	0.6634				
	9	0.0135	0.0007	0.0017	0.2651	0.6363	0.0231	0.0006	0.0030	0.1448	0.6634				
586	3	0.899	1.14	33.70	0.1063	15.15	-0.002	0.09	-0.000	0.1293	0.6492				
	8	0.1018	0.0021	0.0119	0.2317	0.5844	0.0245	0.0006	0.0031	0.1382	0.6537				
	9	0.0194	0.0009	0.0025	0.2317	0.6519	0.0204	0.0005	0.0026	0.1382	0.6537				
586	4	0.898	1.14	45.00	0.1072	15.15	-0.004	0.09	-0.000	0.1320	0.6442				
	8	0.1039	0.0022	0.0122	0.2134	0.5874	0.0187	0.0004	0.0024	0.1248	0.6491				
	9	0.0246	0.0010	0.0032	0.2134	0.6649	0.0165	0.0004	0.0021	0.1248	0.6491				
586	5	0.900	1.14	56.20	0.1159	15.16	-0.005	0.11	-0.001	0.1363	0.6492				
	8	0.1030	0.0023	0.0121	0.1784	0.5894	0.0130	0.0003	0.0016	0.1095	0.6603				
	9	0.0301	0.0010	0.0037	0.1784	0.6201	0.0118	0.0002	0.0015	0.1095	0.6603				
586	6	0.900	1.15	67.49	0.1211	15.16	-0.005	0.11	-0.001	0.1178	0.6522				
	8	0.1022	0.0024	0.0120	0.1771	0.5905	0.0078	0.0001	0.0010	0.0876	0.5748				
	9	0.0326	0.0011	0.0042	0.1771	0.6516	0.0066	0.0001	0.0007	0.0876	0.5748				

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key															
586	1 6 9	0.899 0.1026 0.0344	3.15 0.0023 0.0011	78.69 0.0118 0.0044	0.1160 0.1671	15.15 0.5786 0.6471	-0.0054 0.0029	0.11 0.0000 -0.0001	-0.0004 0.0002	-0.0724 -0.2773	0.5871 0.3450						
586	8 6 9	0.897 0.1021 0.0339	3.15 0.0023 0.0011	89.99 0.0117 0.0044	0.1151 0.1693	15.16 0.5765 0.6527	-0.0009 -0.0031	0.11 0.0003 -0.0003	-0.0003 -0.0006	1.7755 0.5701	1.3477 1.0548						
587	1 7 9	0.900 0.1016 0.0133	6.24 0.0018 0.0008	11.20 0.0122 0.0017	0.0927 0.3270	15.14 0.6022 0.6620	0.02 0.0656 0.0575	0.08 0.0014 0.0018	0.04 0.0086 0.0072	0.1106 0.1571	0.6551 0.6314						
587	2 7 9	0.900 0.1086 0.0263	6.24 0.0020 0.0013	22.50 0.0130 0.0032	0.0940 0.2522	15.15 0.5985 0.6177	0.00 0.0603 0.0527	0.10 0.0012 0.0017	0.03 0.0079 0.0066	0.1063 0.1659	0.6602 0.6325						
587	3 7 9	0.898 0.1145 0.0380	6.24 0.0020 0.0017	33.70 0.0135 0.0049	0.0882 0.2308	15.16 0.5894 0.6508	0.00 0.0520 0.0454	0.12 0.0011 0.0016	0.03 0.0068 0.0058	0.1082 0.1797	0.6616 0.6448						
587	4 7 9	0.899 0.1138 0.0472	6.25 0.0022 0.0020	45.00 0.0131 0.0058	0.0971 0.2171	15.17 0.5794 0.6189	-0.00 0.0410 0.0357	0.14 0.0009 0.0013	0.02 0.0053 0.0045	0.1181 0.1873	0.6554 0.6400						
587	5 7 9	0.898 0.1159 0.0525	6.25 0.0020 0.0023	56.20 0.0133 0.0066	0.0872 0.2216	15.16 0.5761 0.6313	-0.02 0.0298 0.0254	0.15 0.0007 0.0009	0.00 0.0038 0.0031	0.1211 0.1837	0.6376 0.6267						
587	6 7 9	0.900 0.1126 0.0583	6.25 0.0020 0.0022	67.49 0.0131 0.0071	0.0908 0.1947	15.16 0.5828 0.6088	-0.04 0.0181 0.0149	0.16 0.0004 0.0004	-0.00 0.0022 0.0018	0.1314 0.1568	0.6242 0.6219						
587	7 7 9	0.899 0.1116 0.0599	6.25 0.0018 0.0023	78.69 0.0128 0.0076	0.0840 0.1920	15.16 0.5748 0.6402	-0.05 0.0076 0.0050	0.16 0.0001 0.0000	-0.02 0.0009 0.0004	0.1040 0.0350	0.5902 0.4811						
587	8 7 9	0.899 0.1115 0.0617	6.25 0.0018 0.0021	78.69 0.0127 0.0074	0.0825 0.1776	15.16 0.5734 0.6058	-0.05 0.0076 0.0049	0.16 0.0001 0.0000	-0.01 0.0009 0.0004	0.1027 0.0280	0.5903 0.4977						
587	9 7 9	0.900 0.1074 0.0630	6.25 0.0020 0.0020	89.99 0.0126 0.0074	0.0960 0.1601	15.15 0.5869 0.5894	-0.06 -0.0018 -0.0048	0.16 0.0002 -0.0004	-0.04 -0.0003 -0.0009	0.6942 0.4332	0.9591 0.9340						

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key											
588	1	7	0.900	9.35	11.20	0.717	15.15	0.06	0.09	0.06	0.0118	0.1097	0.6271	
		8	0.1093	0.0015	0.0132	0.2930	0.6074	0.0946	0.0020	0.0118	0.0094	0.1439	0.5866	
		9	0.0188	0.0011	0.0023	0.0023	0.6363	0.0806	0.0023	0.0094	0.0094	0.1439	0.5866	
588	2	7	0.900	9.37	22.51	0.786	15.16	0.05	0.12	0.06	0.0113	0.0997	0.6351	
		8	0.1182	0.0018	0.0141	0.2513	0.5989	0.0894	0.0017	0.0113	0.0088	0.1598	0.5899	
		9	0.0368	0.0018	0.0045	0.0045	0.6242	0.0752	0.0024	0.0088	0.0088	0.1598	0.5899	
588	3	7	0.899	9.36	33.71	0.834	15.17	0.03	0.15	0.05	0.0102	0.0959	0.6400	
		8	0.1234	0.0020	0.0144	0.2358	0.5869	0.0794	0.0015	0.0102	0.0080	0.1805	0.5999	
		9	0.0512	0.0024	0.0060	0.0060	0.5894	0.0671	0.0024	0.0080	0.0080	0.1805	0.5999	
588	4	7	0.899	9.36	45.01	0.747	15.17	0.02	0.17	0.04	0.0084	0.0987	0.6544	
		8	0.1261	0.0018	0.0145	0.2152	0.5785	0.0643	0.0012	0.0084	0.0068	0.1928	0.6101	
		9	0.0629	0.0027	0.0073	0.0073	0.5869	0.0560	0.0021	0.0068	0.0068	0.1928	0.6101	
588	5	7	0.899	9.35	56.20	0.726	15.16	0.00	0.19	0.02	0.0059	0.1152	0.6460	
		8	0.1209	0.0017	0.0139	0.2005	0.5761	0.0460	0.0010	0.0059	0.0054	0.1981	0.6355	
		9	0.0716	0.0028	0.0084	0.0084	0.5929	0.0428	0.0016	0.0054	0.0054	0.1981	0.6355	
588	6	7	0.899	9.35	67.49	0.766	15.16	0.02	0.20	0.00	0.0033	0.1281	0.6208	
		8	0.1174	0.0017	0.0135	0.1814	0.5753	0.0273	0.0007	0.0033	0.0032	0.1593	0.6287	
		9	0.0790	0.0028	0.0093	0.0093	0.5887	0.0261	0.0010	0.0032	0.0032	0.1593	0.6287	
588	7	7	0.900	9.36	78.69	0.805	15.16	0.05	0.20	0.01	0.0012	0.1216	0.5757	
		8	0.1149	0.0018	0.0133	0.1597	0.5784	0.0109	0.0002	0.0012	0.0011	0.1802	0.6198	
		9	0.0851	0.0027	0.0098	0.0098	0.5768	0.0093	0.0003	0.0011	0.0011	0.1802	0.6198	
588	8	7	0.897	9.36	89.99	0.724	15.15	0.07	0.20	0.02	0.0005	0.4589	0.9232	
		8	0.1159	0.0016	0.0131	0.1575	0.5670	0.0030	0.0002	0.0005	0.0005	0.4860	1.1047	
		9	0.0861	0.0027	0.0105	0.0105	0.6126	0.0041	0.0004	0.0009	0.0009	0.4860	1.1047	
589	1	7	0.901	12.50	11.20	0.622	15.15	0.07	0.10	0.07	0.0138	0.1018	0.6045	
		8	0.1151	0.0014	0.0143	0.2793	0.6226	0.1144	0.0023	0.0138	0.0110	0.1146	0.5678	
		9	0.0234	0.0013	0.0028	0.0028	0.6010	0.0974	0.0022	0.0110	0.0110	0.1146	0.5678	
589	2	7	0.899	12.50	22.51	0.618	15.17	0.07	0.14	0.08	0.0135	0.1032	0.6142	
		8	0.1266	0.0015	0.0151	0.2697	0.5977	0.1106	0.0022	0.0135	0.0104	0.1287	0.5654	
		9	0.0445	0.0024	0.0054	0.0054	0.6137	0.0926	0.0023	0.0104	0.0104	0.1287	0.5654	
589	3	7	0.899	12.49	33.71	0.707	15.18	0.06	0.18	0.07	0.0127	0.0925	0.6291	
		8	0.1332	0.0018	0.0156	0.2337	0.5867	0.1015	0.0018	0.0127	0.0096	0.1499	0.5730	
		9	0.0629	0.0029	0.0072	0.0072	0.5777	0.0844	0.0025	0.0096	0.0096	0.1499	0.5730	

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key →															
589	4	7	0.899	12.48	45.01	0.0632	15.18	0.09	0.20	0.06	0.0871	0.6512						
		8	0.1313	0.0016	0.0151	0.2007	0.5767	0.0847	0.0014	0.0110	0.1810	0.5834						
		9	0.0770	0.0030	0.0089	0.0000	0.5786	0.0725	0.0026	0.0084	0.0000	0.0000						
589	5	7	0.901	12.46	56.20	0.0842	15.17	0.01	0.21	0.05	0.0963	0.6522						
		8	0.1220	0.0020	0.0142	0.1660	0.5859	0.0609	0.0011	0.0079	0.2050	0.6018						
		9	0.0883	0.0029	0.0098	0.0000	0.5555	0.0569	0.0023	0.0068	0.0000	0.0000						
589	6	7	0.899	12.47	67.49	0.0723	15.16	-0.01	0.21	0.02	0.1140	0.6326						
		8	0.1194	0.0017	0.0137	0.1439	0.5748	0.0363	0.0008	0.0045	0.2056	0.6293						
		9	0.0980	0.0028	0.0110	0.0000	0.5626	0.0387	0.0015	0.0048	0.0000	0.0000						
589	7	7	0.899	12.48	78.69	0.0678	15.16	-0.04	0.23	-0.00	0.1405	0.5986						
		8	0.1202	0.0016	0.0139	0.1281	0.5795	0.0135	0.0003	0.0016	0.1836	0.6035						
		9	0.1065	0.0027	0.0120	0.0000	0.5640	0.0159	0.0005	0.0019	0.0000	0.0000						
589	8	7	0.898	12.48	90.00	0.0620	15.16	-0.07	0.22	-0.03	0.3149	0.7971						
		8	0.1206	0.0014	0.0137	0.1267	0.5717	0.0447	0.0002	0.0007	0.6043	1.1763						
		9	0.1094	0.0027	0.0128	0.0000	0.5885	0.0034	0.0004	0.0008	0.0000	0.0000						
590	3	7	1.250	-0.003	-0.00	0.0458	2.04	-1.84	-2.02	2.00	0.1791	0.7544						
		8	0.0122	0.0001	0.0017	0.0120	0.7162	-0.0056	0.0002	0.0008	0.0421	0.6763						
		9	-0.0100	-0.0000	-0.0013	0.0000	0.6905	0.0112	0.0000	0.0015	0.0000	0.0000						
590	4	7	1.249	-0.003	-0.00	-0.0436	1.01	-0.93	-1.02	0.96	0.7596	1.2750						
		8	0.0071	-0.0000	0.0009	-0.0854	0.6872	-0.0007	0.0001	0.0001	-0.0189	0.7006						
		9	-0.0051	0.0000	-0.0007	0.0000	0.7634	0.0058	0.0000	0.0000	0.0000	0.0000						
591	1	7	1.250	4.20	-0.00	0.0225	2.04	2.16	-2.01	6.07	0.0655	0.6721						
		8	0.0116	0.0000	0.0015	-0.0376	0.6846	0.0527	0.0006	0.0070	0.0370	0.6582						
		9	-0.0102	0.0000	-0.0013	0.0000	0.6575	0.0721	0.0005	0.0094	0.0000	0.0000						
591	2	7	1.250	6.31	-0.00	-0.0084	2.03	2.19	-2.01	6.09	0.0610	0.6627						
		8	0.0110	-0.0000	0.0015	-0.0426	0.6851	0.0701	0.0008	0.0093	0.0364	0.6439						
		9	-0.0099	0.0000	-0.0012	0.0000	0.6325	0.0895	0.0006	0.0115	0.0000	0.0000						
591	3	7	1.249	7.44	-0.00	-0.0077	2.04	5.12	-2.01	9.10	0.0585	0.6444						
		8	0.0109	-0.0000	0.0015	-0.0267	0.7041	0.0928	0.0010	0.0119	0.0368	0.6235						
		9	-0.0097	0.0000	-0.0013	0.0000	0.6831	0.1101	0.0008	0.0137	0.0000	0.0000						
591	4	7	1.251	5.29	-0.00	-0.0174	2.03	5.09	-2.01	9.09	0.0597	0.6584						
		8	0.0112	-0.0000	0.0015	-0.0462	0.6687	0.0765	0.0009	0.0100	0.0364	0.6398						
		9	-0.0100	0.0000	-0.0012	0.0000	0.6314	0.0951	0.0006	0.0121	0.0000	0.0000						

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key															
591	5	7 8 9	1.249 0.0114 -0.0102	7.48 -0.0000 0.0000	-0.0015 0.0014 -0.0014	-0.0132 -0.0134 0.0134	2.05 0.6901 0.6951	5.14 0.1059 0.1203	-2.01 0.0012 0.0010	9.13 0.0133 0.0146	0.0578 0.0423 0.0423	0.6297 0.6104 0.6104						
592	1	7 8 9	1.251 0.0060 -0.0053	7.42 -0.0001 0.0002	-0.0008 0.0006 -0.0006	-0.0838 -0.2224 0.2224	1.01 0.6920 0.5610	8.19 0.1050 0.1137	-1.03 0.0011 0.0008	10.13 0.0132 0.0141	0.0569 0.0376 0.0376	0.6328 0.6209 0.6209						
592	2	7 8 9	1.251 0.0058 -0.0055	9.48 -0.0000 0.0001	-0.0008 0.0007 -0.0008	-0.0830 -0.1440 0.1440	1.01 0.6770 0.7325	8.21 0.1165 0.1220	-1.03 0.0013 0.0011	10.14 0.0144 0.0147	0.0575 0.0463 0.0463	0.6204 0.6041 0.6041						
593	1	7 8 9	1.251 0.0118 -0.0090	4.22 -0.0000 0.0000	-0.0015 0.0014 -0.0014	-0.0098 0.0396 0.0396	2.03 0.6536 0.8092	3.72 0.0612 0.0826	-2.02 0.0008 0.0006	8.06 0.0082 0.0108	0.0680 0.0393 0.0393	0.6769 0.6550 0.6550						
593	2	7 8 9	1.250 0.0113 -0.0088	6.32 -0.0000 0.0000	-0.0015 0.0013 -0.0013	-0.0312 0.0191 0.0191	2.03 0.6699 0.7670	3.74 0.0777 0.0990	-2.02 0.0009 0.0007	8.10 0.0102 0.0126	0.0537 0.0567 0.0567	0.6616 0.6391 0.6391						
593	3	7 8 9	1.251 0.0113 -0.0100	8.44 -0.0000 0.0000	-0.0015 0.0013 -0.0013	-0.0244 0.0235 0.0235	2.04 0.6842 0.6848	3.77 0.0934 0.1126	-2.02 0.0011 0.0008	8.11 0.0120 0.0140	0.0609 0.0371 0.0371	0.6452 0.6238 0.6238						
594	1	7 8 9	0.900 0.0086 -0.0046	-0.003 0.0002 -0.0001	-0.0007 0.0011 -0.0007	0.1287 0.1205 0.1205	1.01 0.6885 0.7710	-0.94 0.0003 0.0063	-1.02 0.0001 0.0001	0.95 0.0000 0.0008	-2.3960 -0.1308 -0.1308	-0.3535 -0.6957 -0.6957						
594	2	7 8 9	0.900 0.0150 -0.0118	-0.003 0.0001 -0.0001	-0.0020 0.0017 -0.0017	0.1116 0.0511 0.0511	2.04 0.6793 0.7376	-1.84 0.0051 -0.0138	-2.01 0.0003 0.0001	2.02 0.0008 0.0018	0.3024 0.0709 0.0709	0.8160 0.6778 0.6778						
595	1	7 8 9	0.899 0.0146 -0.0085	5.23 0.0002 -0.0002	-0.0019 0.0015 -0.0015	0.0748 0.1391 0.1391	2.04 0.6568 0.8946	3.07 0.0747 0.0888	-2.02 0.0021 0.0021	7.09 0.0092 0.0104	0.1441 0.1232 0.1232	0.6181 0.5904 0.5904						
595	2	7 8 9	0.902 0.0160 -0.0108	3.13 0.0002 -0.0001	-0.0019 0.0015 -0.0015	0.0644 0.0568 0.0568	2.04 0.6146 0.7322	3.05 0.0586 0.0740	-2.02 0.0016 0.0018	7.07 0.0090 0.0090	0.1374 0.1236 0.1236	0.6453 0.6127 0.6127						
596	1	7 8 9	0.901 0.0086 -0.0036	8.33 -0.0000 -0.0000	-0.0010 0.0010 -0.0006	-0.0328 0.0995 0.0995	1.01 0.6259 0.8496	8.17 0.1099 0.1144	-1.03 0.0025 0.0018	10.10 0.0128 0.0151	0.1145 0.0800 0.0800	0.5831 0.5731 0.5731						

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key													
596	2	7	0.900	6.28	-0.00	0.0157	1.01	0.6522	0.1021	8.17	-1.03	10.10	0.1271	0.5904	
		8	0.0081	0.0000	0.0010	0.0303	0.6522	0.1021	0.0025	0.0120	0.0025	0.0120	0.1016	0.5725	
		9	-0.0033	-0.0000	-0.0006	0.0303	0.9195	0.1065	0.0021	0.0121	0.0021	0.0121			
597	1	7	0.900	6.27	-0.00	0.0381	2.04	0.6179	5.12	-2.02	9.08	0.1362	0.5969		
		8	0.0151	0.0001	0.0018	0.0649	0.6179	0.0905	0.0024	0.0108	0.0024	0.0108	0.1061	0.5769	
		9	-0.0093	-0.0001	-0.0012	0.0649	0.6825	0.1031	0.0021	0.0119	0.0021	0.0119			
597	2	7	0.900	8.35	-0.00	0.0370	2.04	0.6268	5.14	-2.02	9.08	0.1235	0.5895		
		8	0.0149	0.0001	0.0018	0.1284	0.6268	0.1045	0.0025	0.0123	0.0025	0.0123	0.0876	0.5724	
		9	-0.0086	-0.0002	-0.0012	0.1284	0.7241	0.1122	0.0019	0.0128	0.0019	0.0128			
597	3	7	0.901	4.18	-0.00	0.0645	2.04	0.6289	5.10	-2.02	9.07	0.1481	0.6097		
		8	0.0156	0.0002	0.0019	0.0520	0.6289	0.0756	0.0022	0.0092	0.0022	0.0092	0.1233	0.5870	
		9	-0.0105	-0.0001	-0.0015	0.0520	0.7090	0.0892	0.0022	0.0104	0.0022	0.0104			
598	1	7	0.900	3.14	-0.00	0.0877	2.05	0.6386	4.06	-2.02	8.06	0.1407	0.6310		
		8	0.0156	0.0002	0.0020	0.0509	0.6386	0.0637	0.0017	0.0080	0.0017	0.0080	0.1253	0.6022	
		9	-0.0110	-0.0001	-0.0015	0.0509	0.7063	0.0780	0.0019	0.0094	0.0019	0.0094			
598	2	7	0.900	5.23	-0.00	0.0563	2.04	0.6354	4.09	-2.02	8.08	0.1445	0.6092		
		8	0.0150	0.0001	0.0019	0.1046	0.6354	0.0786	0.0022	0.0095	0.0022	0.0095	0.1191	0.5849	
		9	-0.0090	-0.0001	-0.0014	0.1046	0.8205	0.0925	0.0022	0.0108	0.0022	0.0108			
598	3	7	0.899	7.32	-0.00	0.0478	2.04	0.6362	4.10	-2.02	8.08	0.1345	0.5978		
		8	0.0147	0.0001	0.0018	0.0478	0.6362	0.0928	0.0024	0.0110	0.0024	0.0110	0.1040	0.5699	
		9	-0.0085	-0.0002	-0.0012	0.0478	0.7571	0.1050	0.0021	0.0119	0.0021	0.0119			
599	3	7	0.798	-3.13	-0.00	-0.2499	-0.02	0.7514	-3.03	0.05	-3.03	0.2622	0.6305		
		8	0.0018	-0.0000	0.0002	-0.8609	0.7514	-0.0416	-0.0021	-0.0052	-0.0021	-0.0052	0.2510	0.6440	
		9	-0.0011	0.0001	-0.0002	-0.8609	1.0512	-0.0419	-0.0021	-0.0054	-0.0021	-0.0054			
599	4	7	0.801	-1.10	-0.00	0.0770	-0.02	0.9518	-2.98	0.06	-2.99	0.2733	0.6752		
		8	0.0024	0.0000	0.0004	0.0770	0.9518	-0.0210	-0.0011	-0.0028	-0.0011	-0.0028	0.2958	0.6706	
		9	-0.0006	0.0002	-0.0001	-1.8987	1.3868	-0.0208	-0.0012	-0.0028	-0.0012	-0.0028			
599	5	7	0.801	-0.03	-0.00	0.0654	-0.02	0.7700	-2.97	0.06	-2.97	0.2781	0.7175		
		8	0.0032	0.0000	0.0004	0.0654	0.7700	-0.0116	-0.0006	-0.0016	-0.0006	-0.0016	0.3442	0.7048	
		9	-0.0006	0.0002	-0.0001	-1.8660	0.7335	-0.0121	-0.0008	-0.0017	-0.0008	-0.0017			
599	6	7	0.797	0.46	-0.00	-0.0008	-0.02	0.6596	-2.95	0.06	-2.97	0.3183	0.8198		
		8	0.0038	-0.0000	0.0005	-0.0008	0.6596	-0.0067	-0.0004	-0.0010	-0.0004	-0.0010	0.3927	0.7363	
		9	-0.0006	0.0002	-0.0000	-2.0782	0.7113	-0.0080	-0.0006	-0.0011	-0.0006	-0.0011			

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key															
599	7	0.798	0.0000	0.98	-0.00	0.0581	-0.02	-2.74	0.06	-2.96	0.4004	1.1166						
	8	0.0038	0.0000	0.0000	0.0005	0.0581	-0.02	-2.74	0.06	-2.96	0.4004	1.1166						
	9	-0.0010	0.0002	-0.0000	-0.0000	-1.4057	0.2915	-0.0040	-0.0004	-0.0006	0.5235	0.7924						
599	8	0.800	0.0000	0.12	-0.00	-0.0167	-0.02	-2.91	0.06	-2.94	0.2474	0.6185						
	8	0.0034	-0.0000	0.0004	0.0004	-0.0167	-0.02	-2.91	0.06	-2.94	0.2474	0.6185						
	9	-0.0008	0.0002	-0.0000	-0.0000	-1.5937	0.3376	0.0107	0.0003	0.0013	0.1719	0.6191						
599	9	0.798	0.0000	0.19	-0.00	-0.2599	-0.02	-2.85	0.06	-2.88	0.2469	0.6178						
	8	0.0024	-0.0001	0.0002	0.0002	-0.2599	-0.02	-2.85	0.06	-2.88	0.2469	0.6178						
	9	0.0010	0.0002	-0.0000	-0.0000	1.3345	-0.2768	0.0415	0.0017	0.0052	0.2043	0.6319						
599	10	0.796	0.0000	0.30	-0.00	-0.2124	-0.02	-2.80	0.06	-2.84	0.2535	0.5697						
	8	0.0021	-0.0000	0.0002	0.0002	-0.2124	-0.02	-2.80	0.06	-2.84	0.2535	0.5697						
	9	0.0001	0.0003	0.0000	0.0000	11.3896	2.9646	0.0659	0.0028	0.0076	0.2121	0.5773						
599	11	0.797	0.0000	0.48	-0.00	-0.1443	-0.01	-2.79	0.06	-2.82	0.1804	0.5502						
	8	0.0019	-0.0000	0.0003	0.0003	-0.1443	-0.01	-2.79	0.06	-2.82	0.1804	0.5502						
	9	-0.0006	0.0002	-0.0000	-0.0000	-2.1136	0.3509	0.0834	0.0025	0.0092	0.1544	0.5521						
599	12	0.799	0.0000	0.06	-0.00	-0.0188	-0.01	-2.97	0.06	-2.97	0.3040	0.7557						
	8	0.0037	-0.0000	0.0004	0.0004	-0.0188	-0.01	-2.97	0.06	-2.97	0.3040	0.7557						
	9	-0.0009	0.0003	-0.0000	-0.0000	-1.5612	0.4281	-0.0121	-0.0008	-0.0017	0.3390	0.7129						
600	1	0.797	0.0000	0.10	-0.00	-0.2635	-0.02	-1.01	0.05	-1.06	0.2742	0.6539						
	8	0.0019	-0.0001	0.0002	0.0002	-0.2635	-0.02	-1.01	0.05	-1.06	0.2742	0.6539						
	9	-0.0019	0.0002	-0.0001	-0.0001	-0.7476	0.3385	-0.0294	-0.0015	-0.0038	0.2703	0.6584						
600	2	0.799	0.0000	0.10	-0.00	-0.0507	-0.01	-0.97	0.06	-1.02	0.2895	0.7341						
	8	0.0029	-0.0000	0.0004	0.0004	-0.0507	-0.01	-0.97	0.06	-1.02	0.2895	0.7341						
	9	-0.0009	0.0002	-0.0001	-0.0001	-1.4136	0.7691	-0.0097	-0.0007	-0.0013	0.3638	0.7106						
600	3	0.799	0.0000	0.07	-0.00	0.0367	-0.01	-0.95	0.06	-1.01	0.9293	1.8748						
	8	0.0035	0.0000	0.0005	0.0005	0.0367	-0.01	-0.95	0.06	-1.01	0.9293	1.8748						
	9	-0.0006	0.0002	-0.0001	-0.0001	-2.1300	0.9677	-0.0020	-0.0003	-0.0003	0.8137	0.8822						
600	4	0.799	0.0000	0.49	-0.00	0.0632	-0.01	-0.95	0.06	-1.00	0.2388	0.4956						
	8	0.0037	0.0000	0.0005	0.0005	0.0632	-0.01	-0.95	0.06	-1.00	0.2388	0.4956						
	9	-0.0004	0.0002	-0.0001	-0.0001	-2.6959	1.1414	0.0009	-0.0001	0.0000	-0.6154	0.4358						
600	5	0.800	0.0000	0.98	-0.00	-0.0140	-0.01	-0.94	0.06	-1.00	0.2892	0.6129						
	8	0.0041	-0.0000	0.0005	0.0005	-0.0140	-0.01	-0.94	0.06	-1.00	0.2892	0.6129						
	9	-0.0009	0.0002	-0.0000	-0.0000	-1.5498	0.3246	0.0041	0.0000	0.0005	0.0659	0.6385						

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key														
600	6	0.799 0.0036 -0.0004	5.09 -0.0000 0.0002	-0.00 0.0004 -0.0001	-0.0335 -2.9119	-0.01 0.5564 1.3243	-0.097 0.0276 0.0223	0.006 0.0009	-0.96 0.0034 0.0028	0.2507 0.2047	0.6210 0.6386					
600	7	0.798 0.0020 0.0003	6.20 -0.0000 0.0003	-0.00 0.0003 0.0000	-0.0664 4.5122	-0.01 0.8394 0.5637	-0.84 0.0523 0.0527	0.06 0.0026 0.0021	-0.91 0.0062 0.0064	0.2551 0.2081	0.6003 0.6112					
600	8	0.795 0.0020 -0.0001	9.33 -0.0000 0.0003	-0.00 0.0003 0.0001	-0.1735 -9.4410	-0.01 0.7407 -2.7178	-0.81 0.0736 0.0751	0.06 0.0033 0.0028	-0.88 0.0082 0.0084	0.2256 0.1915	0.5609 0.5656					
600	9	0.797 0.0022 -0.0009	12.50 -0.0000 0.0002	-0.00 0.0002 -0.0000	-0.2244 -1.6291	-0.02 0.6423 0.1634	-0.81 0.0857 0.0870	0.06 0.0026 0.0022	-0.88 0.0094 0.0096	0.1517 0.1269	0.5519 0.5523					
600	10	0.799 0.0035 -0.0010	-0.05 0.0000 0.0003	-0.00 0.0005 -0.0000	0.0350 -1.4963	-0.01 0.7259 0.2093	-0.96 -0.0003 -0.0019	0.06 0.0001 -0.0003	-1.01 0.0002 -0.0003	1.9377 0.8709	3.4877 1.0251					
601	1	0.797 0.0020 -0.0011	-3.11 -0.0000 0.0002	-0.00 0.0002 -0.0002	-0.2294 -1.1035	-0.01 0.7250 0.9403	-0.11 -0.0233	0.06 -0.0013 -0.0013	-0.07 -0.0031 -0.0030	0.2836 0.2888	0.6600 0.6562					
601	2	0.798 0.0031 -0.0008	-1.06 -0.0000 0.0002	-0.00 0.0004 -0.0001	-0.0825 -1.6936	-0.01 0.684 0.9557	-0.08 -0.0037 -0.0042	0.06 0.0002 -0.0004	-0.03 -0.0006 -0.0006	0.3768 0.5194	0.8597 0.7553					
601	3	0.798 0.0035 -0.0004	-0.04 0.0000 0.0002	-0.00 0.0005 -0.0001	0.0267 -2.8714	-0.01 0.7141 1.3660	-0.06 0.0034 0.0019	0.06 0.0001 -0.0000	-0.02 0.0003 0.0002	0.2324 -0.1651	0.5354 0.6293					
601	4	0.798 0.0041 -0.0007	0.49 -0.0000 0.0002	-0.00 0.0004 -0.0000	-0.0495 -1.9807	-0.01 0.5773 0.6642	-0.05 0.0075 0.0052	0.06 0.0004 0.0001	-0.01 0.0009 0.0007	0.2688 0.1459	0.6167 0.7249					
601	5	0.798 0.0043 -0.0004	1.00 -0.0000 0.0002	-0.00 0.0004 -0.0001	-0.0463 -3.0105	-0.01 0.5612 1.2425	-0.03 0.0124 0.0095	0.06 0.0005 0.0003	-0.00 0.0016 0.0012	0.2386 0.1650	0.6472 0.6527					
601	6	0.798 0.0037 0.0002	3.11 -0.0000 0.0002	-0.00 0.0003 -0.0001	-0.0994 3.9716	-0.01 0.4889 -3.7593	0.00 0.0338 0.0282	0.06 0.0016 0.0011	0.02 0.0042 0.0036	0.2436 0.2077	0.6213 0.6434					

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key													
601	7	0.799	6.22	-0.002	-0.000	-0.1905	-0.01	0.06	0.06	0.0028	0.0066	0.2551	0.5819		
	6	0.0021	-0.0000	0.0000	0.0000	1.5106	0.6905	0.0568	0.0024	0.0066	0.0066	0.2146	0.5999		
	9	0.0009	0.0002	-0.0000	-0.0000	-0.2629	-0.0572	0.0572	0.0024	0.0066	0.0066	0.2146	0.5999		
601	8	0.797	9.33	-0.002	-0.000	-0.1999	-0.01	0.08	0.06	0.0033	0.0086	0.2170	0.5570		
	6	0.0021	-0.0000	0.0000	0.0000	6.8965	0.6895	0.0772	0.0033	0.0086	0.0086	0.1874	0.5613		
	9	0.0002	0.0003	0.0000	0.0000	-0.9578	0.0781	0.0781	0.0029	0.0087	0.0087	0.1874	0.5613		
601	9	0.797	12.50	-0.002	-0.000	-0.2359	-0.01	0.08	0.06	0.0024	0.0095	0.1399	0.5519		
	6	0.0022	-0.0001	0.0000	0.0000	-1.2072	0.6201	0.0866	0.0024	0.0095	0.0095	0.1161	0.5553		
	9	-0.0013	0.0003	0.0000	0.0000	-0.0666	0.0883	0.0883	0.0020	0.0098	0.0098	0.1161	0.5553		
601	10	0.798	12.50	-0.002	-0.000	-0.2674	-0.01	0.07	0.05	0.0024	0.0095	0.1393	0.5509		
	6	0.0022	-0.0001	0.0000	0.0000	-1.6879	0.5758	0.0867	0.0024	0.0095	0.0095	0.1160	0.5547		
	9	-0.0008	0.0002	-0.0000	-0.0000	0.3421	0.0883	0.0883	0.0020	0.0098	0.0098	0.1160	0.5547		
601	11	0.799	-0.04	-0.004	-0.000	-0.0032	-0.01	-0.05	0.06	0.0001	-0.003	0.1767	0.4876		
	6	0.0036	-0.0000	0.0004	0.0004	-1.7017	0.6709	0.0037	0.0001	0.0001	0.0002	-0.1337	0.5104		
	9	-0.0008	0.0002	-0.0001	-0.0001	0.6298	0.0020	0.0020	-0.0000	0.0000	0.0002	-0.1337	0.5104		
602	1	0.798	-3.09	-0.002	-0.000	-0.2859	-0.02	0.96	0.05	0.0008	0.0022	0.2670	0.6554		
	6	0.0022	-0.0001	0.0002	0.0002	-0.7501	0.7951	-0.0164	-0.0010	-0.0020	-0.0020	0.3134	0.6374		
	9	-0.0011	0.0001	-0.0001	-0.0001	0.7446	0.7446	0.0012	0.0000	0.0000	0.0000	0.2670	0.6374		
602	2	0.798	-1.06	-0.004	-0.000	-0.0447	-0.02	1.01	0.05	0.0001	0.0094	0.2699	0.5444		
	6	0.0030	-0.0000	0.0004	0.0004	-2.3257	1.7433	0.0006	-0.0001	-0.0001	0.0000	-1.2079	0.6553		
	9	-0.0003	0.0001	-0.0001	-0.0001	0.7446	0.7446	0.0006	0.0000	0.0000	0.0000	-1.2079	0.6553		
602	3	0.797	-0.04	-0.005	-0.000	0.0105	-0.02	1.03	0.06	0.0004	0.0096	0.2885	0.6780		
	6	0.0037	0.0000	0.0005	0.0005	-1.8155	0.7347	0.0082	0.0002	0.0002	0.0011	0.1672	0.7425		
	9	-0.0006	0.0002	-0.0002	-0.0002	0.3417	0.3417	0.0066	0.0002	0.0002	0.0009	0.1672	0.7425		
602	4	0.797	0.52	-0.005	-0.000	0.0477	-0.02	1.04	0.05	0.0007	0.0018	0.2542	0.6639		
	6	0.0036	0.0000	0.0005	0.0005	-3.7314	1.7259	0.0138	0.0004	0.0004	0.0015	0.1839	0.6686		
	9	-0.0002	0.0002	-0.0000	-0.0000	0.7259	0.7259	0.0112	0.0004	0.0004	0.0015	0.1839	0.6686		
602	5	0.796	1.01	-0.005	-0.000	0.0820	-0.02	1.05	0.06	0.0009	0.0024	0.2577	0.6476		
	6	0.0035	0.0000	0.0005	0.0005	-1.7265	0.7265	0.0191	0.0006	0.0006	0.0020	0.2016	0.6625		
	9	-0.0007	0.0002	-0.0000	-0.0000	0.3630	0.3630	0.0155	0.0006	0.0006	0.0020	0.2016	0.6625		
602	6	0.798	3.12	-0.003	-0.000	-0.0994	-0.02	1.10	0.05	0.0020	0.0050	0.2504	0.6266		
	6	0.0038	-0.0000	0.0003	0.0003	-2.5804	1.1568	0.0399	0.0014	0.0014	0.0046	0.2002	0.6440		
	9	-0.0004	0.0002	-0.0001	-0.0001	0.4889	0.4889	0.0358	0.0014	0.0014	0.0046	0.2002	0.6440		

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key									
602	7 8 9	0.799 0.0022 0.0002	6.22 -0.0000 0.0002	-0.0003 0.0000 0.0000	-0.1630 5.1622	-0.02 0.7220 0.8704	1.14 0.0599 0.0619	0.06 0.0030 0.0026	1.05 0.0069 0.0073	0.2583 0.2122	0.5775 0.5895
602	7 8 9	0.797 0.0019 0.0000	9.35 -0.0000 0.0002	-0.0003 0.0000 0.0000	-0.1574 33.0932	-0.02 0.8279 7.0355	1.16 0.0781 0.0807	0.06 0.0032 0.0028	1.07 0.0086 0.0090	0.2100 0.1745	0.5542 0.5587
602	7 8 9	0.797 0.0029 -0.0019	12.51 -0.0002 0.0003	-0.0002 0.0001 0.0001	-0.3420 -0.9008	-0.02 0.3910 -0.3190	1.15 0.0874 0.0895	0.05 0.0023 0.0019	1.06 0.0097 0.0099	0.1322 0.1076	0.5560 0.5558
602	7 8 9	0.798 0.0032 -0.0007	-0.04 0.0000 0.0002	-0.0005 0.0000 -0.0000	0.0598 -1.7365	-0.02 0.8014 0.3924	1.03 0.0085 0.0065	0.06 0.0004 0.0002	0.96 0.0010 0.0008	0.2526 0.1552	0.6269 0.6746
603	7 8 9	0.797 0.0019 -0.0012	-3.09 -0.0001 0.0002	-0.0002 0.0001 -0.0001	-0.2623 -0.9413	-0.02 0.7435 0.6542	2.95 -0.0050 -0.0048	0.05 -0.0003 -0.0004	3.03 -0.0007 -0.0006	0.3500 0.4721	0.7827 0.6786
603	7 8 9	0.798 0.0026 -0.0005	-1.03 0.0000 0.0002	-0.0004 0.0001 -0.0001	0.0626 -2.2035	-0.02 0.9133 0.9850	2.99 0.0114 0.0103	0.06 0.0005 0.0003	3.07 0.0015 0.0013	0.2355 0.1865	0.6739 0.6664
603	7 8 9	0.798 0.0038 -0.0006	-0.01 0.0000 0.0002	-0.0005 0.0000 -0.0000	-0.0088 -2.0777	-0.02 0.6771 0.4355	3.01 0.0211 0.0187	0.06 0.0010 0.0007	3.08 0.0027 0.0024	0.2485 0.2019	0.6475 0.6481
603	7 8 9	0.798 0.0037 -0.0012	0.52 0.0000 0.0003	-0.0005 0.0000 -0.0000	0.0534 -1.2596	-0.02 0.7275 0.0060	3.02 0.0271 0.0236	0.06 0.0013 0.0009	3.09 0.0034 0.0030	0.2451 0.2085	0.6371 0.6528
603	7 8 9	0.798 0.0037 -0.0008	1.05 0.0000 0.0002	-0.0005 0.0000 -0.0000	0.0448 -1.6359	-0.02 0.6949 0.1748	3.03 0.0322 0.0289	0.06 0.0015 0.0011	3.10 0.0041 0.0037	0.2461 0.2004	0.6386 0.6445
603	7 8 9	0.799 0.0030 -0.0003	3.14 0.0000 0.0002	-0.0004 0.0000 -0.0000	0.0285 -3.8239	-0.02 0.7176 1.5701	3.07 0.0505 0.0485	0.06 0.0025 0.0019	3.13 0.0061 0.0061	0.2550 0.2040	0.6053 0.6302
603	7 8 9	0.799 0.0023 0.0003	6.23 -0.0001 0.0003	-0.0002 0.0001 0.0000	-0.2189 -4.9068	-0.02 0.6213 0.7499	3.10 0.0698 0.0709	0.06 0.0031 0.0027	3.17 0.0079 0.0081	0.2285 0.1924	0.5690 0.5735

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key											
603	8	7 8 9	0.797 0.0022 0.0001	2.54 -0.0000 0.0002	-0.00 0.0003 0.0000	-0.2029 12.1719	-0.02 0.6847 2.2258	3.11 0.0831 0.0850	0.06 0.0028 0.0024	3.18 0.0092 0.0094	0.1685 0.1415	0.5534 0.5548		
603	9	7 8 9	0.798 0.0021 -0.0006	12.51 -0.0000 0.0002	-0.00 0.0003 -0.0000	-0.1668 -2.3410	-0.01 0.7531 0.0468	3.11 0.0899 0.0919	0.06 0.0022 0.0019	3.17 0.0100 0.0101	0.1228 0.1035	0.5566 0.5545		
603	10	7 8 9	0.796 0.0037 -0.0006	0.00 -0.0000 0.0002	-0.00 0.0004 -0.0000	-0.0127 -2.2479	-0.02 0.6571 0.7463	3.00 0.216 0.0191	0.06 0.0010 0.0008	3.09 0.0027 0.0024	0.2440 0.2093	0.6383 0.6511		
604	1	7 8 9	0.797 0.0018 -0.0009	-3.08 -0.0000 0.0002	-0.00 0.0003 -0.0001	-0.2490 -1.1013	-0.02 0.7932 0.9568	6.01 0.0103 0.0091	0.05 0.0004 0.0003	6.00 0.0014 0.0012	0.2184 0.1874	0.6847 0.6975		
604	2	7 8 9	0.798 0.0027 -0.0007	-1.03 0.0000 0.0002	-0.00 0.0004 -0.0001	0.0157 -1.8263	-0.02 0.8335 0.8345	6.06 0.0309 0.0287	0.05 0.0014 0.0011	6.05 0.0040 0.0037	0.2414 0.2066	0.6487 0.6590		
604	3	7 8 9	0.797 0.0038 -0.0008	-0.00 -0.0000 0.0002	-0.00 0.0004 -0.0000	-0.0312 -1.5937	-0.02 0.6305 0.3376	6.08 0.0405 0.0387	0.05 0.0019 0.0015	6.05 0.0051 0.0050	0.2409 0.1979	0.6320 0.6481		
604	4	7 8 9	0.798 0.0039 -0.0008	0.54 -0.0000 0.0002	-0.00 0.0005 -0.0000	-0.0113 -1.7072	-0.02 0.6355 0.2755	6.09 0.0449 0.0434	0.06 0.0022 0.0017	6.07 0.0055 0.0055	0.2472 0.1988	0.6220 0.6416		
604	5	7 8 9	0.798 0.0038 -0.0007	1.04 0.0000 0.0002	-0.00 0.0005 -0.0000	0.0124 -1.9171	-0.02 0.6738 0.4617	6.10 0.0490 0.0476	0.06 0.0024 0.0019	6.08 0.0060 0.0060	0.2528 0.2009	0.6132 0.6335		
604	6	7 8 9	0.798 0.0033 -0.0006	3.15 -0.0000 0.0002	-0.00 0.0004 -0.0000	-0.0542 -2.0777	-0.02 0.6036 0.4355	6.12 0.0624 0.0622	0.06 0.0030 0.0025	6.11 0.0071 0.0072	0.2445 0.2080	0.5734 0.5867		
604	7	7 8 9	0.797 0.0023 0.0004	6.25 -0.0000 0.0003	-0.00 0.0003 0.0000	-0.1705 3.9437	-0.02 0.6514 0.7256	6.14 0.0795 0.0814	0.06 0.0032 0.0028	6.12 0.0088 0.0091	0.2023 0.1719	0.5540 0.5586		
604	8	7 8 9	0.796 0.0022 0.0005	9.38 -0.0000 0.0002	-0.00 0.0003 0.0000	-0.1703 2.5200	-0.02 0.7463 0.4384	6.13 0.0858 0.0884	0.06 0.0022 0.0019	6.10 0.0095 0.0098	0.1324 0.1080	0.5559 0.5543		

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key →																
604	7	799	12.53	-0.000	-0.00	-0.02	6.14	0.06	6.11	0.1127	0.5569								
	8	0:0020	-0.0000	0.0003	0.7718	0.0943	0.0021	0.0105	0.0952	0.0953	0.5555								
	9	-0.0009	0.0002	-0.0000	0.0090	(0.952	0.0018	0.0105											
604	7	797	0.00	-0.00	-0.01	0.07	0.06	6.06	0.2357	0.6247									
	8	0:0038	-0.0000	0.0004	0.6550	0.0407	0.0019	0.0050	0.0386	0.2010	0.6519								
	9	-0.0013	0.0003	-0.0000	0.0153	0.0386	0.0015	0.0050											
605	7	797	-3.04	-0.00	-0.02	9.09	0.05	9.03	0.2333	0.6574									
	8	0:0019	0.0000	0.0003	0.8397	0.0293	0.0013	0.0038	0.0275	0.2047	0.6702								
	9	-0.0013	0.0002	-0.0001	0.4880	0.0275	0.0011	0.0036											
605	7	797	-0.99	-0.00	-0.02	9.13	0.06	9.06	0.2487	0.6248									
	8	0:0031	-0.0000	0.0004	0.6924	0.0474	0.0023	0.0059	0.0476	0.2000	0.6354								
	9	-0.0011	0.0003	-0.0000	0.1759	0.0476	0.0019	0.0060											
605	7	797	0.02	-0.00	-0.02	9.15	0.06	9.07	0.2575	0.5982									
	8	0:0033	0.0000	0.0005	0.8011	0.0544	0.0028	0.0065	0.0544	0.2089	0.6108								
	9	-0.0007	0.0002	-0.0000	0.4103	0.0544	0.0022	0.0066											
605	7	799	0.58	-0.00	-0.02	9.15	0.06	9.09	0.2540	0.5867									
	8	0:0038	0.0000	0.0005	0.6707	0.0577	0.0029	0.0067	0.0580	0.2084	0.6049								
	9	-0.0006	0.0002	-0.0000	0.4927	0.0580	0.0024	0.0070											
605	7	798	1.07	-0.00	-0.02	9.16	0.06	9.09	0.2447	0.5783									
	8	0:0036	0.0000	0.0005	0.6896	0.0611	0.0029	0.0070	0.0613	0.2066	0.5939								
	9	-0.0003	0.0002	-0.0000	1.2433	0.0613	0.0025	0.0072											
605	7	797	3.17	-0.00	-0.02	9.18	0.06	9.10	0.2111	0.5680									
	8	0:0029	0.0000	0.0004	0.7304	0.0766	0.0032	0.0087	0.0766	0.1801	0.5768								
	9	-0.0006	0.0002	-0.0000	0.4761	0.0754	0.0027	0.0087											
605	7	798	6.29	-0.00	-0.02	9.18	0.06	9.09	0.1541	0.5556									
	8	0:0022	-0.0000	0.0003	0.7288	0.0845	0.0026	0.0093	0.0845	0.1305	0.5547								
	9	0:0007	0.0002	-0.0000	-0.0705	0.0861	0.0022	0.0095											
605	7	797	9.37	-0.00	-0.02	9.17	0.06	9.09	0.1165	0.5614									
	8	0:0019	-0.0000	0.0003	0.8682	0.0902	0.0021	0.0101	0.0902	0.0953	0.5609								
	9	0:0002	0.0002	0.0000	2.1148	0.0923	0.0017	0.0103											
605	7	797	12.56	-0.00	-0.02	9.18	0.06	9.10	0.1044	0.5605									
	8	0:0022	-0.0001	0.0003	0.6901	0.0978	0.0020	0.0109	0.0978	0.0859	0.5652								
	9	-0.0007	0.0002	-0.0000	0.1033	0.0991	0.0017	0.0110											

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key →										y, 08	0:2577	0:2071	0:5982 0:6075		
		0:798 0:0035 -0:0009	0:005 0:0002	-0:005 -0:0000	0:0237 -1:5038	-0:002 0:0545 0:0545	0:005 0:0000	0:0002 -0:0000	0:002 -0:0000	0:0237 -1:5038	-0:002 0:0545 0:0545						
605	10	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	
606	1	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	
606	2	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	
606	3	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	
606	4	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	
606	5	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	
606	6	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	
606	7	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	
606	8	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	
606	9	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	
606	10	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	
		0:796 0:0020 -0:0011	0:005 -0:0001 0:0002	-0:005 0:0002 -0:0001	0:0237 -1:5038 -0:9472	-0:002 0:0545 0:0545	0:005 0:0000 0:0002	0:0002 -0:0000 -0:0001	0:0237 -1:5038 -0:9472	-0:002 0:0545 0:0545	0:005 0:0022 0:0018	0:005 0:0057 0:0059	0:005 0:0029 0:0073	12:07 12:00 12:09	0:2449 0:2008 0:2050	0:2449 0:2008 0:2050	0:6358 0:6438 0:5877 0:5967
		0:797 0:0023 -0:0004	0:000 0:0002 -0:0002	-0:000 0:0005 -0:0001	0:2000 -2:6366	12:09 12:09 0:0612	0:000 0:0005 -0:0001	0:2000 -2:6366	12:09 12:09 0:0612	0:06 0:0029 0:0025	0:06 0:0029 0:0025	12:09 0:0069 0:0073	0:06 0:0029 0:0025	12:10 12:10 12:11	0:2231 0:1914	0:2231 0:1914	0:5810 0:5902
		0:797 0:0037 -0:0006	0:000 0:0002 0:0002	-0:000 0:0005 -0:0000	-0:0242 -1:9923	12:11 0:0672 0:0675	0:000 0:0005 -0:0000	-0:0242 -1:9923	12:11 0:0672 0:0675	0:06 0:0030 0:0025	0:06 0:0030 0:0025	12:11 0:0078 0:0079	0:06 0:0030 0:0025	12:11 12:11 12:11	0:2159 0:1796	0:2159 0:1796	0:5785 0:5829
		0:799 0:0035 -0:0008	0:000 0:0002 0:0002	-0:000 0:0005 -0:0000	0:0454 -1:5447	12:11 0:0754 0:0749	0:000 0:0005 -0:0000	0:0454 -1:5447	12:11 0:0754 0:0749	0:06 0:0031 0:0026	0:06 0:0031 0:0026	12:11 0:0086 0:0087	0:06 0:0031 0:0026	12:11 12:11 12:11	0:2089 0:1762	0:2089 0:1762	0:5739 0:5823
		0:799 0:0030 -0:0007	0:000 0:0002 0:0002	-0:000 0:0005 -0:0000	-0:0318 -1:8518	12:12 0:0813 0:0834	0:000 0:0005 -0:0000	-0:0318 -1:8518	12:12 0:0813 0:0834	0:06 0:0029 0:0026	0:06 0:0029 0:0026	12:11 12:10 12:10	0:06 0:0029 0:0026	12:11 12:10 12:10	0:1810 0:1565	0:1810 0:1565	0:5549 0:5593
		0:797 0:0021 0:0002	0:000 0:0003 0:0003	-0:000 0:0000 0:0000	-0:1912 3:7976	12:11 0:0869 0:0890	0:000 0:0003 0:0000	-0:1912 3:7976	12:11 0:0869 0:0890	0:06 0:0021 0:0018	0:06 0:0021 0:0018	12:10 0:0097 0:0099	0:06 0:0021 0:0018	12:10 12:10 12:10	0:1261 0:1037	0:1261 0:1037	0:5587 0:5609
		0:797 0:0018 0:0000	0:000 0:0002 0:0002	-0:000 0:0003 0:0000	-0:1609 4:3338	12:12 0:0946 0:0969	0:000 0:0003 0:0000	-0:1609 4:3338	12:12 0:0946 0:0969	0:06 0:0021 0:0017	0:06 0:0021 0:0017	12:10 0:0106 0:0108	0:06 0:0021 0:0017	12:10 12:10 12:10	0:1149 0:0893	0:1149 0:0893	0:5599 0:5600
		0:798 0:0019 -0:0004	0:000 0:0002 0:0002	-0:000 0:0003 -0:0000	-0:2003 -3:0225	12:12 0:1009 0:1028	0:000 0:0003 -0:0000	-0:2003 -3:0225	12:12 0:1009 0:1028	0:06 0:0019 0:0016	0:06 0:0019 0:0016	12:10 0:0113 0:0115	0:06 0:0019 0:0016	12:10 12:10 12:10	0:0936 0:0780	0:0936 0:0780	0:5633 0:5599
		0:797 0:0033 -0:0007	0:001 0:0002 0:0002	-0:001 0:0005 -0:0000	0:0583 -1:8853	12:10 0:0673 0:0675	0:001 0:0005 -0:0000	0:0583 -1:8853	12:10 0:0673 0:0675	0:05 0:0029 0:0025	0:05 0:0029 0:0025	12:10 0:0077 0:0079	0:05 0:0029 0:0025	12:10 12:10 12:10	0:2223 0:1910	0:2223 0:1910	0:5773 0:5877

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd																						
607	1	7	0.746	-3.00	-0.00	-0.00	-0.02	15.12	0.05	15.10	0.2433	0.5927	0.0021	-0.0000	0.0003	-0.1970	0.8019	0.575	0.0028	15.10	0.0068	0.2023	0.6016
		8	0.0010	0.0002	-0.0001	0.0001	0.8071	0.0608	0.0024	0.0073													
		9	-0.0010	0.0002	-0.0001	0.0001	0.8071	0.0608	0.0024	0.0073													
607	2	7	0.798	-0.96	-0.00	-0.01	15.14	0.06	15.11	0.2108	0.5797	0.0033	0.0033	0.0002	0.0002	-0.17249	0.6481	0.0737	0.0031	15.11	0.0085	0.1705	0.5827
		8	0.0033	-0.0000	0.0004	0.6552	0.0754	0.0025	0.0087														
		9	-0.0008	0.0002	-0.0001	0.6481	0.0754	0.0025	0.0087														
607	3	7	0.797	0.02	-0.00	-0.01	15.15	0.06	15.11	0.2009	0.5695	0.0033	0.0033	0.0002	0.0002	0.0533	0.8078	0.0799	0.0032	15.11	0.0091	0.1618	0.5731
		8	0.0033	0.0000	0.0005	0.8078	0.0808	0.0026	0.0092														
		9	-0.0005	0.0002	-0.0000	0.7056	0.0808	0.0026	0.0092														
607	4	7	0.797	0.59	-0.00	-0.02	15.15	0.06	15.12	0.1957	0.5626	0.0036	0.0036	0.0002	0.0002	0.0266	0.7546	0.0818	0.0032	15.12	0.0092	0.1593	0.5671
		8	0.0036	0.0000	0.0005	0.7546	0.0829	0.0026	0.0094														
		9	-0.0004	0.0002	-0.0000	0.8617	0.0829	0.0026	0.0094														
607	5	7	0.798	1.10	-0.00	-0.01	15.15	0.06	15.12	0.1889	0.5570	0.0036	0.0036	0.0002	0.0002	0.0435	0.7481	0.0817	0.0030	15.12	0.0091	0.1542	0.5616
		8	0.0036	0.0000	0.0005	0.7481	0.0834	0.0025	0.0093														
		9	-0.0004	0.0002	-0.0000	0.8727	0.0834	0.0025	0.0093														
607	6	7	0.799	3.22	-0.00	-0.01	15.14	0.06	15.12	0.1438	0.5554	0.0036	0.0036	0.0002	0.0002	-0.0195	0.7155	0.0856	0.0024	15.12	0.0095	0.1215	0.5575
		8	0.0036	-0.0000	0.0004	0.7155	0.0880	0.0021	0.0098														
		9	-0.0007	0.0002	-0.0000	0.3158	0.0880	0.0021	0.0098														
607	7	7	0.799	6.29	-0.00	-0.01	15.14	0.05	15.11	0.1190	0.5600	0.0036	0.0036	0.0002	0.0002	0.0975	0.9243	0.0936	0.0021	15.11	0.0105	0.0958	0.5637
		8	0.0019	-0.0000	0.0003	0.9243	0.0936	0.0017	0.0105														
		9	0.0010	0.0002	-0.0000	0.3158	0.0936	0.0017	0.0105														
607	8	7	0.798	9.37	-0.00	-0.01	15.14	0.05	15.12	0.1039	0.5603	0.0024	0.0024	0.0002	0.0002	0.2199	0.7109	0.1013	0.0020	15.12	0.0111	0.0830	0.5603
		8	0.0024	-0.0001	0.0003	0.7109	0.1013	0.0016	0.0111														
		9	0.0005	0.0002	0.0000	0.2749	0.1013	0.0016	0.0111														
607	9	7	0.797	12.58	-0.00	-0.02	15.15	0.06	15.12	0.0903	0.5603	0.0021	0.0021	0.0002	0.0002	0.2079	0.8497	0.1081	0.0015	15.12	0.0120	0.0700	0.5580
		8	0.0021	-0.0000	0.0003	0.8497	0.1081	0.0015	0.0120														
		9	-0.0007	0.0002	-0.0000	0.1494	0.1081	0.0015	0.0120														
607	10	7	0.798	0.05	-0.00	-0.01	15.15	0.06	15.12	0.2000	0.5600	0.0037	0.0037	0.0003	0.0003	-0.0380	0.6703	0.0800	0.0032	15.12	0.0090	0.1604	0.5600
		8	0.0037	-0.0000	0.0005	0.6703	0.0811	0.0026	0.0092														
		9	-0.0012	0.0003	-0.0000	0.0729	0.0811	0.0026	0.0092														
608	3	7	0.601	-3.10	-0.00	-0.02	15.15	0.05	15.12	0.2674	0.6322	0.0015	0.0015	0.0001	0.0001	-0.2578	0.7396	0.0381	0.0020	15.12	0.0048	0.2673	0.6482
		8	0.0015	-0.0000	0.0002	0.7396	0.0377	0.0020	0.0048														
		9	-0.0016	0.0001	-0.0001	0.4701	0.0377	0.0020	0.0048														

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key											
608	4	7	0.601	-1.09	-0.00	0.1625	-0.02	-2.98	0.05	-2.98	0.2699	0.6522	
		6	0.0024	0.0000	0.0004	0.1625	0.9288	-0.0190	-0.0010	-0.0024	0.5084	0.6701	
		9	-0.0007	0.0001	-0.0001	-0.9454	1.0751	-0.0193	-0.0011	-0.0025			
608	5	7	0.601	-0.05	-0.00	0.1348	-0.02	-2.96	0.05	-2.97	0.2573	0.6915	
		6	0.0030	0.0000	0.0005	0.1348	0.8242	-0.0098	-0.0005	-0.0013	0.3467	0.6917	
		9	-0.0008	0.0001	-0.0001	-1.0092	0.6347	-0.0109	-0.0007	-0.0015			
608	6	7	0.600	0.48	-0.00	0.1683	-0.02	-2.95	0.05	-2.96	0.2508	0.7612	
		6	0.0030	0.0001	0.0005	0.1683	0.8512	-0.0056	-0.0002	-0.0008	0.3984	0.7290	
		9	-0.0006	0.0001	-0.0001	-1.1918	1.0345	-0.0070	-0.0005	-0.0010			
608	7	7	0.599	0.98	-0.00	0.0760	-0.02	-2.94	0.05	-2.95	0.2368	1.0694	
		6	0.0034	0.0000	0.0004	0.0760	0.6785	-0.0017	-0.0000	-0.0003	0.5221	0.8389	
		9	-0.0009	0.0001	-0.0001	-0.9419	0.6440	-0.0036	-0.0003	-0.0006			
608	8	7	0.600	3.10	-0.00	0.1400	-0.02	-2.91	0.05	-2.93	0.3030	0.6297	
		6	0.0024	0.0000	0.0004	0.1400	0.8304	0.0128	0.0007	0.0016	0.1905	0.5995	
		9	-0.0002	0.0001	-0.0001	-2.3616	3.0176	0.0099	0.0003	0.0011			
608	9	7	0.600	6.18	-0.00	0.1784	-0.02	-2.86	0.05	-2.88	0.2772	0.6102	
		6	0.0022	-0.0000	0.0002	0.1784	0.5491	0.0386	0.0021	0.0047	0.2258	0.6176	
		9	0.0002	0.0002	0.0000	4.0657	0.8151	0.0384	0.0017	0.0047			
608	10	7	0.600	9.27	-0.00	0.2233	-0.02	-2.81	0.05	-2.84	0.2770	0.5737	
		6	0.0018	-0.0000	0.0002	0.2233	0.6004	0.0617	0.0034	0.0070	0.2336	0.5811	
		9	0.0001	0.0001	0.0000	5.3312	1.3864	0.0638	0.0029	0.0074			
608	11	7	0.600	12.44	-0.00	0.2655	-0.02	-2.79	0.05	-2.82	0.2366	0.5467	
		6	0.0020	-0.0001	0.0002	0.2655	0.5632	0.0792	0.0037	0.0086	0.2083	0.5490	
		9	-0.0009	0.0001	-0.0000	-0.9497	0.1000	0.0827	0.0034	0.0090			
608	12	7	0.600	-0.04	-0.00	0.1656	-0.02	-2.96	0.05	-2.96	0.2675	0.7046	
		6	0.0028	0.0000	0.0004	0.1656	0.8791	-0.0098	-0.0005	-0.0013	0.3513	0.7145	
		9	-0.0011	0.0001	-0.0000	-0.8372	0.4024	-0.0107	-0.0007	-0.0015			
609	1	7	0.600	-3.09	-0.00	0.3162	-0.03	-1.01	0.05	-1.03	0.2725	0.6347	
		6	0.0019	-0.0001	0.0002	0.3162	0.5499	-0.0267	-0.0014	-0.0033	0.2878	0.6569	
		9	-0.0016	0.0001	-0.0001	-0.4791	0.4414	-0.0263	-0.0015	-0.0034			
609	2	7	0.600	-1.09	-0.00	0.0310	-0.02	-0.96	0.05	-1.01	0.2827	0.7238	
		6	0.0030	-0.0000	0.0004	0.0310	0.6657	-0.0076	-0.0004	-0.0011	0.3817	0.7001	
		9	-0.0007	0.0001	-0.0001	-1.0173	0.9317	-0.0084	-0.0006	-0.0011			

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key													
609	5	7	0.600	-0.000	-0.005	0.1374	-0.002	-0.004	0.000	0.04	-0.001	-0.1125	1.6494			
		8	0.0031	0.0000	0.0005	-0.8364	0.8095	-0.0004	0.0000	-0.0001	-0.0003	0.6876	0.9169			
		9	-0.0010	0.0001	-0.0000	-0.1374	0.4370	-0.0020	-0.0002	0.0000	-0.0003	-0.6876	0.9169			
609	4	7	0.600	0.48	-0.005	0.1152	-0.02	-0.029	0.002	0.05	-0.003	0.4058	0.5653			
		8	0.0034	0.0000	0.0005	-1.0630	0.7522	0.0029	0.0002	0.0000	0.0000	-0.4058	0.1224			
		9	-0.0006	0.0001	-0.0001	1.0630	0.9728	0.0011	-0.0001	0.0000	0.0000	-0.4058	0.1224			
609	5	7	0.600	0.99	-0.005	0.0604	-0.02	-0.093	0.005	0.05	-0.008	0.3582	0.6540			
		8	0.0035	0.0000	0.0004	-0.7087	0.6604	0.0064	0.0004	0.0000	0.0004	0.1031	0.6073			
		9	-0.0011	0.0001	-0.0000	0.7087	0.3614	0.0038	0.0000	0.0000	0.0004	0.1031	0.6073			
609	6	7	0.600	3.12	-0.005	0.1493	-0.02	-0.089	0.005	0.05	-0.030	0.2942	0.6246			
		8	0.0025	0.0000	0.0004	-0.7259	0.8289	0.0242	0.0014	0.0008	0.0024	0.2174	0.6093			
		9	-0.0012	0.0001	-0.0000	0.7259	0.2082	0.0205	0.0008	0.0000	0.0024	0.2174	0.6093			
609	7	7	0.600	6.20	-0.005	0.1956	-0.02	-0.084	0.005	0.04	-0.091	0.2834	0.5980			
		8	0.0021	-0.0000	0.0002	-3.9011	0.5429	0.0492	0.0027	0.0060	0.0060	0.2286	0.6068			
		9	-0.0002	0.0002	0.0000	3.9011	0.9077	0.0498	0.0022	0.0060	0.0060	0.2286	0.6068			
609	8	7	0.601	9.30	-0.005	0.1158	-0.02	-0.080	0.005	0.05	-0.087	0.2712	0.5658			
		8	0.0015	-0.0000	0.0002	-2.1998	0.7733	0.0702	0.0038	0.0063	0.0063	0.2250	0.5711			
		9	-0.0003	0.0001	0.0000	2.1998	0.2129	0.0733	0.0033	0.0063	0.0063	0.2250	0.5711			
609	9	7	0.600	12.44	-0.005	0.0314	-0.02	-0.080	0.005	0.05	-0.091	0.2080	0.5446			
		8	0.0015	-0.0000	0.0003	-19.5384	0.9612	0.0840	0.0034	0.0094	0.0094	0.1842	0.5395			
		9	-0.0000	0.0001	-0.0001	19.5384	0.1425	0.0871	0.0032	0.0094	0.0094	0.1842	0.5395			
609	10	7	0.600	-0.05	-0.005	0.0164	-0.02	-0.095	0.004	0.04	-1.00	-0.0436	2.0073			
		8	0.0035	0.0000	0.0004	-0.8864	0.6361	0.0004	0.0000	-0.0003	-0.0003	-0.7973	1.1200			
		9	-0.0009	0.0001	-0.0000	0.8864	0.4769	-0.0017	-0.0002	-0.0003	-0.0003	-0.7973	1.1200			
610	1	7	0.601	-3.09	-0.005	0.2668	-0.02	-0.11	0.004	0.04	-0.07	0.2767	0.6306			
		8	0.0020	-0.0001	0.0002	-0.5412	0.5018	-0.0213	-0.0011	-0.0026	-0.0026	0.3062	0.6534			
		9	-0.0018	0.0001	-0.0000	0.5412	0.2126	-0.0205	-0.0012	-0.0026	-0.0026	0.3062	0.6534			
610	2	7	0.601	-1.08	-0.005	0.1616	-0.02	-0.08	0.004	0.04	-0.04	0.2459	0.7834			
		8	0.0024	0.0000	0.0004	-0.9449	0.9107	-0.0034	-0.0004	-0.0005	-0.0005	0.4980	0.7317			
		9	-0.0008	0.0001	-0.0001	0.9449	0.7462	-0.0040	-0.0004	-0.0005	-0.0005	0.4980	0.7317			
610	3	7	0.600	-0.05	-0.005	0.0664	-0.02	-0.06	0.004	0.04	-0.003	0.3994	0.5983			
		8	0.0031	0.0000	0.0004	-0.9597	0.7593	0.0031	0.0002	0.0003	0.0003	-0.1215	0.4148			
		9	-0.0007	0.0001	-0.0001	0.9597	0.6925	-0.0019	-0.0000	-0.0003	-0.0003	-0.1215	0.4148			

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key															
610	4	7	0.599	0.501	-0.005	0.1774	-0.025	-0.069	0.004	0.05	-0.003	0.3469	0.6755					
		8	0.0031	0.0001	0.0005	0.8205	0.0069	0.0004	0.0009	0.0009	0.0009	0.1488	0.6299					
		9	-0.0006	0.0001	-0.0001	-0.9755	0.8808	0.0049	0.0001	0.0006	0.0006							
610	5	7	0.599	0.97	-0.00	0.1401	-0.02	-0.04	0.03	-0.02	0.2961	0.6477						
		8	0.0033	0.0000	0.0004	0.7491	0.0113	0.0006	0.0003	0.0014	0.1750	0.6171						
		9	-0.0004	0.0001	-0.0001	-1.3578	1.2467	0.0086	0.0003	0.0010								
610	6	7	0.600	5.12	-0.00	0.1080	-0.02	-0.04	0.03	0.00	0.2838	0.6233						
		8	0.0027	0.0000	0.0004	0.7738	0.0300	0.0017	0.0017	0.0032	0.2262	0.6264						
		9	-0.0002	0.0001	-0.0001	-2.7064	2.7947	0.0262	0.0011	0.0032								
610	7	7	0.600	6.20	-0.00	0.2084	-0.02	0.04	0.04	0.05	0.2875	0.5906						
		8	0.0021	-0.0000	0.0002	0.5592	0.0531	0.0030	0.0030	0.0062	0.2335	0.6017						
		9	0.0004	0.0002	0.0000	-2.5829	0.7637	0.0543	0.0025	0.0065								
610	8	7	0.600	9.29	-0.00	0.1364	-0.02	0.08	0.04	0.08	0.2651	0.5553						
		8	0.0016	-0.0000	0.0002	0.7360	0.0732	0.0038	0.0038	0.0081	0.2233	0.5599						
		9	0.0004	0.0001	-0.0000	-1.6026	0.0768	0.0034	0.0034	0.0086								
610	9	7	0.600	12.45	-0.00	0.1354	-0.02	0.07	0.04	0.08	0.1924	0.5443						
		8	0.0013	-0.0000	0.0002	0.9223	0.0856	0.0032	0.0032	0.0093	0.1717	0.5413						
		9	-0.0005	0.0001	-0.0001	-1.0984	0.9795	0.0888	0.0030	0.0096								
610	10	7	0.600	-0.04	-0.00	0.0611	-0.0257	-0.06	0.04	-0.03	0.3671	0.5566						
		8	0.0039	-0.0000	0.0004	-1.2495	1.0082	0.0031	0.0002	0.0001	-0.1280	0.4370						
		9	-0.0005	0.0001	-0.0001	-1.2495	1.0082	0.0016	-0.0000	0.0001								
611	1	7	0.600	-3.08	-0.00	0.2146	-0.02	0.97	0.04	0.92	0.2804	0.6483						
		8	0.0019	-0.0000	0.0002	0.5574	0.0148	-0.0148	0.0008	-0.0019	0.3253	0.6519						
		9	-0.0011	0.0001	-0.0001	-0.6679	0.4973	-0.0147	-0.0009	-0.0019	0.4479	0.4728						
611	2	7	0.600	-1.06	-0.00	0.0305	-0.02	1.01	0.04	0.95	-2.5072	-0.7280						
		8	0.0030	-0.0000	0.0004	0.6798	0.0012	0.0003	0.0001	0.0001	0.3369	0.6960						
		9	-0.0014	0.0001	-0.0000	0.1911	0.0003	-0.0001	-0.0001	-0.0007	0.1434	0.6368						
611	3	7	0.600	-0.04	-0.00	0.1835	-0.02	1.03	0.03	0.95	0.3369	0.6960						
		8	0.0029	0.0001	0.0005	0.8583	0.0078	0.0062	0.0005	0.0010	0.1434	0.6368						
		9	-0.0008	0.0001	-0.0000	0.4812	0.0062	0.0001	0.0001	0.0007	0.3369	0.6368						
611	4	7	0.600	0.50	-0.00	0.0923	-0.02	1.04	0.04	0.96	0.2993	0.6594						
		8	0.0035	0.0000	0.0004	0.6938	0.0124	0.0078	0.0007	0.0016	0.1842	0.6301						
		9	-0.0005	0.0001	-0.0001	-1.3176	1.0212	0.0099	0.0003	0.0012	0.2993	0.6301						

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																		
611	5	0.599	1.00	-0.00	0.0062	-0.02	1.04	0.03	0.97	0.2917	0.6414	0.0037	0.0000	0.0004	0.5751	0.0170	0.0009	0.0021	0.2043	0.6384
	6	-0.0009	0.0001	-0.0001	-0.7827	0.5299	0.0137	0.0005	0.0017	0.2043	0.6384									
611	6	0.600	3.13	-0.00	-0.0126	-0.02	1.09	0.04	1.00	0.2798	0.6224	0.0032	0.0000	0.0003	0.5822	0.0365	0.0020	0.0045	0.2288	0.6385
	7	-0.0003	0.0001	-0.0001	-2.2192	1.6670	0.0321	0.0014	0.0041	0.2288	0.6385									
611	7	0.600	6.21	-0.00	0.0713	-0.02	1.14	0.04	1.04	0.2787	0.5813	0.0015	0.0000	0.0002	0.9709	0.0581	0.0032	0.0067	0.2331	0.5894
	8	-0.0006	0.0002	0.0001	-1.9315	-0.9713	0.0594	0.0027	0.0070	0.2331	0.5894									
611	8	0.599	9.31	-0.00	-0.3293	-0.02	1.16	0.04	1.08	0.2556	0.5517	0.0020	0.0001	0.0001	0.4610	0.0761	0.0038	0.0083	0.2166	0.5522
	9	-0.0013	0.0002	0.0001	-0.9332	-0.5742	0.0796	0.0034	0.0087	0.2166	0.5522									
611	9	0.599	12.44	-0.00	-0.1141	-0.02	1.15	0.04	1.08	0.1762	0.5450	0.0015	0.0000	0.0002	0.8284	0.0870	0.0030	0.0094	0.1575	0.5399
	10	-0.0004	0.0001	-0.0000	-1.5483	0.8830	0.0907	0.0028	0.0097	0.1575	0.5399									
611	10	0.600	-0.05	-0.00	0.0800	-0.02	1.03	0.04	0.96	0.3180	0.6704	0.0031	0.0000	0.0004	0.7787	0.0080	0.0005	0.0010	0.1423	0.6128
	1	-0.0014	0.0001	-0.0000	-0.6699	0.1856	0.0064	0.0001	0.0007	0.1423	0.6128									
612	1	0.599	-3.05	-0.00	-0.1705	-0.02	2.96	0.04	3.04	0.2805	0.7336	0.0019	0.0000	0.0002	0.6814	0.0042	-0.0002	-0.0005	0.5165	0.7196
	2	-0.0015	0.0001	-0.0000	-0.5598	0.1973	-0.0039	-0.0004	-0.0005	0.5165	0.7196									
612	2	0.600	-1.04	-0.00	-0.0489	-0.02	3.00	0.03	3.06	0.2998	0.6818	0.0030	0.0000	0.0003	0.6465	0.0103	0.0006	0.0014	0.1890	0.6320
	3	-0.0010	0.0001	-0.0000	-0.7016	0.4027	0.0093	0.0003	0.0011	0.1890	0.6320									
612	3	0.599	-0.00	-0.00	0.1687	-0.02	3.01	0.04	3.07	0.2914	0.6499	0.0031	0.0000	0.0005	0.8550	0.0194	0.0011	0.0025	0.2161	0.6314
	4	-0.0006	0.0001	-0.0001	-1.0725	0.8917	0.0170	0.0007	0.0021	0.2161	0.6314									
612	4	0.599	0.52	-0.00	-0.1158	-0.02	3.02	0.03	3.08	0.2880	0.6433	0.0034	0.0000	0.0005	0.7650	0.0243	0.0014	0.0031	0.2207	0.6299
	5	-0.0008	0.0001	-0.0000	-0.8361	0.4482	0.0215	0.0009	0.0027	0.2207	0.6299									
612	5	0.599	1.00	-0.00	0.0338	-0.02	3.04	0.04	3.09	0.2835	0.6400	0.0036	0.0000	0.0004	0.6407	0.0292	0.0016	0.0037	0.2173	0.6263
	6	-0.0011	0.0001	-0.0000	-0.7077	0.3077	0.0258	0.0011	0.0032	0.2173	0.6263									

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key →														
612	6	7	0.600	5.15	-0.000	0.0004	0.0275	-0.02	0.6404	0.0472	5.07	0.0027	0.04	5.12	0.2858	0.6090	
		8	0.0032	0.0000	0.0000	0.0004	0.0275	0.6404	0.0472	0.0446		0.0020	0.0055	0.2244	0.6136		
		9	-0.0010	0.0001	-0.0000	-0.0000	-0.8900	0.1393	0.0446			0.0020	0.0055	0.2244	0.6136		
612	7	7	0.599	6.21	-0.000	0.0002	-0.1385	-0.02	0.6726	0.0675	3.10	0.04	3.15	0.2677	0.5744		
		8	0.0018	-0.0000	0.0002	0.0002	-0.1385	0.6726	0.0675	0.0702		0.0036	0.0077	0.2225	0.5749		
		9	-0.0001	0.0002	0.0001	-0.0001	-11.5951	-5.5436	0.0702			0.0031	0.0080	0.2225	0.5749		
612	8	7	0.599	9.30	-0.000	0.0002	-0.1681	-0.02	0.7257	0.0815	3.13	0.04	3.17	0.2246	0.5470		
		8	0.0016	-0.0000	0.0002	0.0002	-0.1681	0.7257	0.0815	0.0849		0.0036	0.0077	0.1954	0.5442		
		9	0.0003	0.0001	-0.0000	-0.0000	2.2004	-0.0244	0.0849			0.0033	0.0092	0.1954	0.5442		
612	9	7	0.600	12.44	-0.000	0.0002	-0.1524	-0.02	0.6044	0.0840	3.11	0.04	3.16	0.1613	0.5464		
		8	0.0012	-0.0000	0.0002	0.0002	-0.1524	1.0044	0.0840	0.0894		0.0027	0.0091	0.1379	0.5396		
		9	-0.0007	0.0001	-0.0000	-0.0000	-0.9727	0.3159	0.0894			0.0024	0.0096	0.1379	0.5396		
612	10	7	0.599	-0.01	-0.000	0.0004	-0.0113	-0.02	0.6044	0.0194	3.02	0.04	3.07	0.2837	0.6372		
		8	0.0036	-0.0000	0.0004	0.0004	-0.0113	0.6044	0.0194	0.0168		0.0011	0.0024	0.2213	0.6393		
		9	-0.0006	0.0001	-0.0001	-0.0001	-1.0473	0.8585	0.0168			0.0007	0.0021	0.2213	0.6393		
613	1	7	0.600	-3.05	-0.000	0.0002	-0.2651	-0.02	0.5641	0.0088	6.02	0.04	5.98	0.3147	0.7313		
		8	0.0020	-0.0001	0.0002	0.0002	-0.2651	0.5641	0.0088	0.0089		0.0005	0.0012	0.1865	0.6460		
		9	-0.0012	0.0001	-0.0001	-0.0001	-0.5079	0.5383	0.0089			0.0003	0.0011	0.1865	0.6460		
613	2	7	0.600	-1.03	-0.000	0.0004	0.0580	-0.02	0.8089	0.0280	6.06	0.04	6.03	0.2783	0.6490		
		8	0.0029	0.0000	0.0004	0.0004	0.0580	0.8089	0.0280	0.0262		0.0015	0.0036	0.2154	0.6379		
		9	-0.0009	0.0001	-0.0001	-0.0001	-0.8325	0.6066	0.0262			0.0011	0.0033	0.2154	0.6379		
613	3	7	0.599	0.02	-0.000	0.0005	0.1082	-0.02	0.7649	0.0370	6.08	0.04	6.04	0.2791	0.6378		
		8	0.0033	0.0000	0.0005	0.0005	0.1082	0.7649	0.0348			0.0020	0.0047	0.2169	0.6370		
		9	-0.0008	0.0001	-0.0000	-0.0000	-0.9578	0.5464	0.0348			0.0015	0.0044	0.2169	0.6370		
613	4	7	0.600	0.54	-0.000	0.0005	0.1097	-0.02	0.7354	0.0415	6.09	0.04	6.04	0.2825	0.6320		
		8	0.0035	0.0000	0.0005	0.0005	0.1097	0.7354	0.0394			0.0023	0.0052	0.2166	0.6302		
		9	-0.0005	0.0001	-0.0001	-0.0001	-1.2089	1.1475	0.0394			0.0017	0.0049	0.2166	0.6302		
613	5	7	0.599	1.04	-0.000	0.0005	0.0626	-0.02	0.7089	0.0458	6.10	0.04	6.05	0.2841	0.6192		
		8	0.0035	0.0000	0.0005	0.0005	0.0626	0.7089	0.0437			0.0026	0.0056	0.2210	0.6260		
		9	-0.0012	0.0001	-0.0000	-0.0000	-0.6853	0.1895	0.0437			0.0019	0.0054	0.2210	0.6260		
613	6	7	0.599	3.16	-0.000	0.0004	0.2001	-0.02	0.8937	0.0609	6.13	0.04	6.08	0.2721	0.5846		
		8	0.0025	0.0001	0.0004	0.0004	0.2001	0.8937	0.0609	0.0598		0.0033	0.0071	0.2272	0.5909		
		9	-0.0008	0.0001	-0.0000	-0.0000	-1.1209	0.1493	0.0598			0.0027	0.0070	0.2272	0.5909		

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd																					
613	7	0.598	6.23	-0.00	-0.02	6.15	0.04	6.12	0.2497	0.5552												
	8	0.0017	-0.0000	0.0002	0.7279	0.0771	0.0038	0.0085	0.2122	0.5525												
	9	0.0007	0.0001	-0.0000	-0.1939	0.0801	0.0034	0.0088														
613	7	0.599	9.33	-0.00	-0.02	6.14	0.04	6.11	0.1757	0.5492												
	8	0.0014	-0.0000	0.0002	0.8551	0.0863	0.0030	0.0094	0.1553	0.5433												
	9	0.0002	0.0001	-0.0000	-0.7804	0.0900	0.0027	0.0097														
613	7	0.599	12.44	-0.00	-0.02	6.13	0.04	6.09	0.1437	0.5478												
	8	0.0015	-0.0000	0.0002	0.8008	0.0858	0.0024	0.0094	0.1192	0.5367												
	9	-0.0007	0.0001	-0.0000	0.5822	0.0881	0.0021	0.0094														
613	7	0.598	-0.00	0.00	-0.02	6.07	0.04	6.04	0.2751	0.6319												
	8	0.0031	0.0000	0.0005	0.8236	0.0371	0.0020	0.0046	0.2183	0.6377												
	9	-0.0006	0.0001	-0.0001	0.9880	0.0347	0.0015	0.0044														
614	1	0.599	-3.04	-0.00	-0.02	9.08	0.04	8.56	0.2768	0.6630												
	8	0.0012	0.0000	0.0003	1.3003	0.0261	0.0014	0.0034	0.2132	0.6480												
	9	-0.0013	0.0001	-0.0001	0.5296	0.0235	0.0010	0.0030														
614	2	0.599	-1.01	-0.00	-0.02	9.12	0.04	8.69	0.2801	0.6285												
	8	0.0032	-0.0000	0.0004	0.6268	0.0444	0.0024	0.0055	0.2179	0.6347												
	9	-0.0015	0.0001	-0.0000	0.1470	0.0419	0.0018	0.0053														
614	3	0.599	-0.00	-0.00	-0.02	9.14	0.04	8.71	0.2843	0.6057												
	8	0.0039	-0.0000	0.0004	0.5347	0.0516	0.0029	0.0062	0.2245	0.6204												
	9	-0.0017	0.0002	-0.0000	0.0533	0.0496	0.0022	0.0061														
614	4	0.599	0.55	-0.00	-0.02	9.14	0.04	8.71	0.2781	0.5969												
	8	0.0039	-0.0000	0.0004	0.5816	0.0555	0.0030	0.0066	0.2254	0.6100												
	9	-0.0011	0.0001	-0.0000	0.3299	0.0534	0.0024	0.0065														
614	5	0.598	-3.05	-0.00	-0.02	9.08	0.04	9.02	0.2766	0.6621												
	8	0.0019	-0.0000	0.0002	0.6446	0.0260	0.0014	0.0034	0.2137	0.6546												
	9	-0.0011	0.0001	-0.0001	0.5408	0.0254	0.0010	0.0033														
614	6	0.599	-1.01	-0.00	-0.02	9.12	0.04	9.05	0.2811	0.6305												
	8	0.0032	-0.0000	0.0003	0.5668	0.0443	0.0024	0.0055	0.2199	0.6329												
	9	-0.0011	0.0001	-0.0001	0.4824	0.0438	0.0019	0.0055														
614	7	0.598	0.00	-0.00	-0.02	9.14	0.04	9.06	0.2861	0.6077												
	8	0.0035	0.0000	0.0004	0.6505	0.0516	0.0029	0.0062	0.2248	0.6150												
	9	-0.0014	0.0001	-0.0000	0.1174	0.0515	0.0023	0.0063														

See Key

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key													
614	8	7	0.599	0.000	0.000	-0.000	0.062	0.696	-0.02	0.556	0.04	0.066	0.279	0.597	
		6	0.0035	0.000	0.000	0.000	0.062	0.696	0.306	0.055	0.003	0.006	0.225	0.607	
		5	-0.0009	0.000	-0.000	-0.000	-0.889	0.306	0.002	0.002	0.002	0.006	0.225	0.607	
614	9	7	0.598	0.000	0.000	-0.000	0.194	0.648	-0.02	0.558	0.04	0.066	0.278	0.597	
		6	0.0036	0.000	0.000	0.000	0.194	0.648	0.089	0.055	0.003	0.006	0.226	0.607	
		5	-0.0014	0.000	-0.000	-0.000	-0.703	0.089	0.002	0.002	0.002	0.006	0.226	0.607	
614	10	7	0.599	1.000	0.000	-0.000	0.060	0.725	-0.02	0.595	0.04	0.070	0.271	0.590	
		6	0.0035	0.000	0.000	0.000	0.060	0.725	0.529	0.058	0.003	0.007	0.226	0.602	
		5	-0.0008	0.000	-0.000	-0.000	-0.892	0.529	0.002	0.002	0.002	0.007	0.226	0.602	
614	11	7	0.598	3.16	0.000	-0.000	0.067	0.745	-0.02	0.734	0.04	0.083	0.260	0.567	
		6	0.0028	0.000	0.000	0.000	0.067	0.745	0.409	0.074	0.003	0.008	0.218	0.575	
		5	-0.0007	0.000	-0.000	-0.000	-1.087	0.409	0.002	0.002	0.002	0.008	0.218	0.575	
614	12	7	0.598	6.24	0.000	-0.000	0.163	0.610	-0.02	0.833	0.04	0.091	0.203	0.548	
		6	0.0022	0.000	0.000	0.000	0.163	0.610	1.365	0.083	0.003	0.009	0.182	0.543	
		5	-0.0003	0.000	-0.000	-0.000	-3.368	1.365	-1.365	0.086	0.003	0.009	0.182	0.543	
614	13	7	0.600	9.32	0.000	-0.000	0.017	0.420	-0.02	0.865	0.04	0.095	0.155	0.540	
		6	0.0012	0.000	0.000	0.000	0.017	0.420	1.190	0.085	0.002	0.009	0.129	0.540	
		5	-0.0001	0.000	-0.000	-0.000	-5.102	1.190	-1.190	0.085	0.002	0.009	0.129	0.540	
614	14	7	0.601	12.45	0.000	-0.000	0.113	0.856	-0.02	0.910	0.04	0.098	0.123	0.555	
		6	0.0015	0.000	0.000	0.000	0.113	0.856	0.356	0.091	0.002	0.009	0.100	0.555	
		5	-0.0009	0.000	-0.000	-0.000	-0.787	0.356	0.002	0.002	0.002	0.009	0.100	0.555	
614	15	7	0.601	0.05	0.000	-0.000	0.137	0.809	-0.02	0.515	0.04	0.063	0.282	0.602	
		6	0.0031	0.000	0.000	0.000	0.137	0.809	0.082	0.051	0.002	0.006	0.227	0.618	
		5	-0.0016	0.000	-0.000	-0.000	-0.668	0.082	0.002	0.002	0.002	0.006	0.227	0.618	
615	1	7	0.600	-3.01	0.000	-0.000	0.202	0.603	-0.02	0.425	0.04	0.054	0.275	0.633	
		6	0.0020	0.000	0.000	0.000	0.202	0.603	0.434	0.043	0.002	0.005	0.217	0.633	
		5	-0.0014	0.000	-0.000	-0.000	-0.524	0.434	0.002	0.002	0.002	0.005	0.217	0.633	
615	2	7	0.601	-0.98	0.000	-0.000	0.136	0.905	-0.02	0.580	0.04	0.069	0.270	0.600	
		6	0.0025	0.000	0.000	0.000	0.136	0.905	0.576	0.059	0.003	0.007	0.221	0.600	
		5	-0.0009	0.000	-0.000	-0.000	-0.833	0.576	0.002	0.002	0.002	0.007	0.221	0.600	
615	3	7	0.601	0.05	0.000	-0.000	0.154	0.858	-0.02	0.654	0.04	0.076	0.260	0.587	
		6	0.0030	0.000	0.000	0.000	0.154	0.858	0.237	0.066	0.003	0.007	0.217	0.593	
		5	-0.0011	0.000	-0.000	-0.000	-0.797	0.237	0.002	0.002	0.002	0.007	0.217	0.593	

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																					
616	5	7	0.601	1.05	-0.00	0.0902	-0.02	15.15	0.04	15.11	0.2338	0.5546	8	0.0035	0.0000	0.0005	0.7303	0.0802	0.0037	0.0089	0.2024	0.5566	
		9	-0.0011	0.0001	-0.0000	-0.7966	0.2916	0.0830	0.0033	0.0092													
616	6	7	0.601	3.15	-0.00	0.1599	-0.02	15.14	0.04	15.11	0.1828	0.5498	8	0.0026	0.0000	0.0004	0.8609	0.0855	0.0031	0.0094	0.1639	0.5487	
		9	-0.0007	0.0001	-0.0000	-1.1715	0.2916	0.0881	0.0028	0.0096													
616	7	7	0.601	6.21	-0.00	0.1308	-0.02	15.13	0.04	15.09	0.1496	0.5482	8	0.0017	-0.0000	0.0002	0.7279	0.0841	0.0025	0.0092	0.1209	0.5461	
		9	-0.0001	0.0002	0.0000	-6.8712	-2.7053	0.0887	0.0021	0.0096													
616	8	7	0.601	9.31	-0.00	0.2119	-0.02	15.13	0.04	15.09	0.1217	0.5551	8	0.0019	-0.0000	0.0002	0.6663	0.0895	0.0021	0.0099	0.0989	0.5494	
		9	-0.0007	0.0002	0.0001	-1.4908	-0.7198	0.0925	0.0018	0.0101													
616	9	7	0.601	12.49	-0.00	0.1946	-0.02	15.13	0.04	15.08	0.1104	0.5574	8	0.0018	-0.0000	0.0002	0.6551	0.0947	0.0020	0.0105	0.0904	0.5525	
		9	-0.0020	0.0002	0.0000	-0.5133	-0.0275	0.0967	0.0017	0.0106													
616	10	7	0.601	0.05	-0.00	0.0015	-0.02	15.14	0.04	15.11	0.2478	0.5654	8	0.0037	-0.0000	0.0004	0.6302	0.0764	0.0037	0.0086	0.2074	0.5664	
		9	-0.0008	0.0001	-0.0000	-1.1333	0.5046	0.0797	0.0033	0.0090													

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																				
617	5	7	1.253	-3.18	-0.000	-0.1730	-0.3964	-3.09	0.05	-3.10	0.3579	0.6013	0.0009	-0.0002	-0.0001	-1.1087	-0.2992	-0.0531	-0.0038	-0.0065	0.3661	0.6123
617	6	7	1.252	-0.06	0.000	-3.3266	-0.7640	-2.99	0.05	-3.00	0.4816	0.5890	-0.0003	0.0002	0.000	-0.6599	-0.5493	-0.0141	-0.0016	-0.0020	0.4717	0.6168
617	7	7	1.251	0.97	-0.00	3.9651	-0.9758	-2.95	0.05	-2.97	0.7338	0.6309	0.0002	0.0002	0.000	3.3240	-0.0065	-0.0004	-0.0008	0.6232	0.6429	
617	8	7	1.253	3.12	-0.00	-4.1268	-0.8469	-2.88	0.05	-2.91	0.3945	0.5585	0.0010	-0.0000	0.000	-0.0565	-0.6216	0.0111	0.0008	0.0012	0.3816	0.5575
617	9	7	1.252	6.26	-0.00	0.3641	-0.8407	-2.78	0.05	-2.82	0.3649	0.5865	0.0028	-0.0000	-0.00	-0.0549	-0.2146	0.0488	0.0033	0.0053	0.3592	0.5763
617	10	7	1.251	9.43	-0.00	0.1676	-0.9230	-2.69	0.05	-2.75	0.3121	0.6107	0.0022	-0.0000	-0.00	0.1676	-0.2924	0.0848	0.0050	0.0102	0.2986	0.6052
617	11	7	1.251	12.67	-0.00	0.6375	-0.9242	-2.62	0.05	-2.69	0.2699	0.6164	0.0008	-0.0001	-0.00	0.6375	-0.2961	0.1198	0.0061	0.0146	0.2557	0.6104
617	12	7	1.253	-0.06	0.000	5.3387	-1.2909	-2.99	0.05	-3.00	0.5140	0.6206	0.0001	0.0001	0.000	5.3387	-0.2941	-0.0140	-0.0017	-0.0021	0.4720	0.6398
618	1	7	1.250	-3.14	-0.00	0.0497	-0.6366	-0.16	0.05	-0.11	0.4125	0.5906	-0.0001	-0.0001	-0.00	0.0497	-0.3597	-0.0309	-0.0025	-0.0036	0.4132	0.5970
618	2	7	1.252	-0.04	0.000	3.1620	-0.1348	-0.06	0.05	-0.03	0.4468	0.4962	0.0002	0.0001	0.000	3.1620	-0.0034	0.0003	0.0003	-0.0001	0.4468	0.4962
618	3	7	1.252	1.02	-0.00	1.7436	-0.2795	-0.02	0.05	0.00	0.4770	0.5405	0.0004	0.0001	-0.00	1.7436	-0.0094	0.0137	0.0010	0.0010	0.4770	0.5405

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key													
618	4	1.252	3.15	-0.000	-0.000	1.0005	-0.02	0.03	0.05	0.05	0.0029	0.0025	0.0040	0.4062	0.5719
	8	0.0004	0.0000	-0.0000	-0.0000	-0.0502	-0.2622	0.0358	0.0029	0.0025	0.0029	0.0025	0.0040	0.4062	0.5719
	9	0.0014	-0.0000	0.0000	0.0000	0.0000	0.2064	0.0309	0.0025	0.0025	0.0025	0.0025	0.0040	0.4062	0.5719
618	5	1.250	6.32	-0.000	-0.000	0.0067	-0.02	0.12	0.05	0.05	0.0047	0.0044	0.14	0.3250	0.6095
	8	-0.0009	-0.0000	-0.0000	-0.0000	0.1607	0.3475	0.0725	0.0047	0.0044	0.0047	0.0044	0.14	0.3250	0.6095
	9	0.0019	0.0000	0.0002	0.0000	0.0000	0.6220	0.0709	0.0044	0.0044	0.0044	0.0044	0.14	0.3250	0.6095
618	6	1.252	9.47	-0.000	-0.000	-0.2592	-0.02	0.19	0.05	0.05	0.0057	0.0057	0.21	0.2768	0.6175
	8	-0.0009	0.0000	-0.0000	-0.0000	-0.1274	0.1606	0.1087	0.0057	0.0057	0.0057	0.0057	0.21	0.2768	0.6175
	9	0.0024	-0.0000	0.0001	0.0000	0.0000	0.2671	0.1080	0.0057	0.0057	0.0057	0.0057	0.21	0.2768	0.6175
618	7	1.251	12.72	-0.000	-0.000	0.2525	-0.02	0.26	0.05	0.05	0.0069	0.0065	0.26	0.2451	0.6152
	8	-0.0011	-0.0000	-0.0001	-0.0000	-0.0766	0.4623	0.1408	0.0069	0.0065	0.0069	0.0065	0.26	0.2451	0.6152
	9	0.0011	0.0000	0.0000	0.0000	0.0000	0.4055	0.1407	0.0065	0.0065	0.0065	0.0065	0.26	0.2451	0.6152
618	8	1.252	-0.05	-0.000	-0.000	2.7045	-0.01	-0.08	0.05	0.05	0.0002	0.0002	0.02	0.3267	0.3760
	8	0.0002	0.0001	0.0000	0.0000	-0.2982	0.2439	0.0035	0.0002	0.0002	0.0002	0.0002	0.02	0.3267	0.3760
	9	0.0008	-0.0000	0.0000	0.0000	0.0000	0.3730	-0.0009	0.0002	0.0002	0.0002	0.0002	0.02	0.3267	0.3760
619	1	1.250	-3.14	-0.000	-0.000	-1.1519	-0.02	0.93	0.05	0.05	-0.0019	-0.0019	0.06	0.4300	0.6052
	8	-0.0013	0.0000	-0.0001	-0.0000	-0.2112	0.5264	-0.0235	-0.0019	-0.0019	-0.0019	-0.0019	0.06	0.4300	0.6052
	9	0.0003	-0.0000	0.0000	0.0000	0.0000	-0.3445	-0.0237	-0.0019	-0.0019	-0.0019	-0.0019	0.06	0.4300	0.6052
619	2	1.252	-0.03	-0.000	-0.000	-2.8115	-0.01	1.03	0.05	0.05	0.0007	0.0007	0.06	0.4883	0.5537
	8	-0.0003	0.0002	0.0000	0.0000	-0.3395	0.4650	0.0098	0.0007	0.0007	0.0007	0.0007	0.06	0.4883	0.5537
	9	0.0015	-0.0001	0.0000	0.0000	0.0000	-0.0053	0.0045	0.0007	0.0007	0.0007	0.0007	0.06	0.4883	0.5537
619	3	1.251	1.01	-0.000	-0.000	22.1006	-0.01	1.07	0.04	0.04	0.0018	0.0015	0.04	0.4534	0.5588
	8	0.0000	0.0002	0.0000	0.0000	-0.2048	7.5456	0.0209	0.0018	0.0015	0.0018	0.0015	0.04	0.4534	0.5588
	9	0.0011	-0.0000	0.0000	0.0000	0.0000	0.1353	0.0152	0.0015	0.0015	0.0015	0.0015	0.04	0.4534	0.5588
619	4	1.251	3.16	-0.000	-0.000	2.3725	-0.02	1.13	0.05	0.05	0.0034	0.0030	1.04	0.3869	0.5837
	8	0.0003	0.0001	-0.0000	-0.0000	0.1461	0.2081	0.0441	0.0034	0.0030	0.0034	0.0030	1.04	0.3869	0.5837
	9	0.0009	0.0000	0.0000	0.0000	0.0000	-0.4180	0.0386	0.0030	0.0030	0.0030	0.0030	1.04	0.3869	0.5837
619	5	1.252	6.33	-0.000	-0.000	0.3160	-0.02	1.21	0.06	0.06	0.0050	0.0047	1.12	0.3095	0.6121
	8	-0.0006	-0.0000	-0.0001	-0.0000	0.0396	0.6288	0.0813	0.0050	0.0047	0.0050	0.0047	1.12	0.3095	0.6121
	9	0.0031	0.0000	0.0001	0.0000	0.0000	0.1719	0.0790	0.0047	0.0047	0.0047	0.0047	1.12	0.3095	0.6121
619	6	1.250	9.50	-0.000	-0.000	-0.1477	-0.02	1.28	0.04	0.04	0.0062	0.0059	1.18	0.2662	0.6183
	8	-0.0008	-0.0000	-0.0000	-0.0000	-0.1345	0.5376	0.1169	0.0062	0.0059	0.0062	0.0059	1.18	0.2662	0.6183
	9	0.0017	-0.0000	0.0001	0.0000	0.0000	0.3315	0.1152	0.0059	0.0059	0.0059	0.0059	1.18	0.2662	0.6183

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key →															
619	7	7	1.250	12.72	-0.000	-0.000	-0.000	0.4579	0.0274	-0.02	1.34	0.05	1.22	0.2374	0.6152			
	8	8	-0.0008	-0.0000	-0.0001	-0.0000	0.0274	0.3794	0.7537	0.1480	0.0070	0.0182	0.2260	0.6085				
619	A	7	1.251	-0.002	-0.000	-0.000	-2.2124	-0.5374	-0.01	1.03	0.04	0.95	0.4591	0.5175				
	8	8	-0.0005	0.0000	0.0000	0.0000	-0.2732	0.1924	0.5374	0.0097	0.0008	0.0010	0.8142	0.4566				
	9	9	-0.0010	-0.0000	0.0000	0.0000	-0.2732	0.1924	0.5374	0.0044	0.0007	0.0004	0.8142	0.4566				
620	1	7	1.250	-3.11	-0.000	-0.000	-0.2291	-0.4270	-0.02	2.95	0.04	3.01	0.3697	0.6397				
	8	8	-0.0013	0.0000	-0.0001	-0.0000	-0.7173	-0.4270	0.4780	-0.0083	-0.0006	-0.0010	0.3796	0.6468				
	9	9	-0.0008	-0.0001	-0.0000	-0.0000	-0.7173	-0.4270	0.4780	-0.0099	-0.0007	-0.0012	0.3796	0.6468				
620	2	7	1.251	-0.001	-0.000	-0.000	5.5532	0.4946	-0.02	3.06	0.04	3.11	0.4769	0.5672				
	8	8	0.0001	0.0001	0.0000	0.0000	5.5532	0.4946	0.4946	0.2228	0.0021	0.0025	0.5133	0.5404				
	9	9	0.0012	-0.0000	0.0000	0.0000	-0.3328	0.0996	0.0996	0.0187	0.0019	0.0020	0.5133	0.5404				
620	3	7	1.251	1.04	-0.000	-0.000	1.2177	0.2989	-0.02	3.09	0.04	3.14	0.4255	0.5764				
	8	8	0.0007	0.0001	0.0000	0.0000	1.2177	0.2989	0.2989	0.353	0.0030	0.0040	0.4461	0.5616				
	9	9	0.0011	-0.0000	0.0000	0.0000	-0.1821	0.1592	0.1592	0.0298	0.0026	0.0033	0.4461	0.5616				
620	4	7	1.251	3.17	-0.000	-0.000	2.4347	0.3785	-0.02	3.16	0.04	3.19	0.3459	0.6013				
	8	8	0.0003	0.0001	0.0000	0.0000	2.4347	0.3785	0.3785	0.611	0.0042	0.0073	0.3434	0.5932				
	9	9	0.0010	0.0000	0.0000	0.0000	0.2086	0.3340	0.3340	0.0556	0.0038	0.0065	0.3434	0.5932				
620	5	7	1.251	6.36	-0.000	-0.000	0.7069	1.8104	-0.02	3.23	0.05	3.26	0.2836	0.6172				
	8	8	-0.0004	-0.0000	-0.0001	-0.0001	0.7069	1.8104	1.8104	0.980	0.0055	0.0121	0.2748	0.6111				
	9	9	-0.0028	0.0000	0.0000	0.0000	0.1399	0.3166	0.3166	0.0960	0.0052	0.0117	0.2748	0.6111				
620	6	7	1.251	9.52	-0.000	-0.000	-0.1787	0.2653	-0.02	3.30	0.04	3.32	0.2485	0.6183				
	8	8	-0.0010	0.0000	-0.0000	-0.0000	-0.1787	0.2653	0.2653	0.1319	0.0065	0.0163	0.2381	0.6107				
	9	9	-0.0024	-0.0000	0.0000	0.0000	-0.1479	0.0848	0.0848	0.1308	0.0062	0.0159	0.2381	0.6107				
620	7	7	1.251	12.73	-0.000	-0.000	0.5046	2.9115	-0.02	3.35	0.05	3.36	0.2246	0.6132				
	8	8	-0.0010	-0.0001	-0.0001	-0.0001	0.5046	2.9115	2.9115	0.1612	0.0072	0.0197	0.2122	0.6056				
	9	9	-0.0002	0.0000	0.0000	0.0000	0.3416	2.9115	2.9115	0.1609	0.0068	0.0195	0.2122	0.6056				
620	8	7	1.251	6.01	-0.000	-0.000	7.4009	0.5291	-0.01	3.06	0.04	3.11	0.4599	0.5485				
	8	8	0.0001	0.0001	0.0000	0.0000	7.4009	0.5291	0.5291	0.232	0.0021	0.0025	0.5106	0.5327				
	9	9	0.0009	-0.0000	0.0000	0.0000	-0.2501	0.2222	0.2222	0.0189	0.0019	0.0020	0.5106	0.5327				
621	1	7	1.251	-3.07	-0.000	-0.000	-0.3120	-0.2473	-0.02	6.03	0.04	6.02	0.6053	0.5217				
	8	8	-0.0017	0.0001	-0.0000	-0.0000	-0.3120	-0.2473	-0.2473	0.098	0.0011	0.0010	0.6053	0.5217				
	9	9	-0.0006	-0.0001	-0.0000	-0.0000	-0.9266	-0.2185	-0.2185	0.0063	0.0011	0.0006	0.6053	0.5217				

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key															
621	2	7 6 9	1.248 0.0006 0.0016	0.00 0.0001 -0.0000	-0.00 -0.0000 -0.0000	0.5531 -0.2715 -0.0162	-0.02 -0.1125 -0.0162	6.13 0.0455 0.0406	0.05 0.0036 0.0047	6.11 0.0054 0.0047	0.3975 0.4152	0.5965 0.5820						
621	3	7 6 9	1.251 0.0010 0.0011	1.07 0.0001 -0.0000	-0.00 0.0000 -0.0000	0.5882 0.2032 -0.0191	-0.01 0.0606 -0.0191	6.17 0.0595 0.0538	0.04 0.0042 0.0038	6.13 0.0072 0.0064	0.3562 0.3574	0.6060 0.5947						
621	4	7 6 9	1.252 0.0002 0.0014	3.23 0.0001 0.0000	-0.00 0.0000 -0.0000	3.4556 0.1103 -0.0101	-0.02 0.2161 -0.0101	6.22 0.0853 0.0804	0.05 0.0051 0.0047	6.18 0.0105 0.0098	0.3033 0.2952	0.6166 0.6113						
621	5	7 6 9	1.251 0.0007 0.0018	6.41 0.0000 0.0001	-0.00 -0.0001 0.0002	0.3290 0.3381 0.6105	-0.02 0.2593 0.6105	6.28 0.1201 0.1187	0.06 0.0061 0.0058	6.24 0.0149 0.0145	0.2575 0.2473	0.6219 0.6142						
621	6	7 6 9	1.250 0.0007 0.0021	9.51 0.0000 -0.0000	-0.00 -0.0000 0.0001	-0.1689 -0.1047 -0.2405	-0.02 0.2465 0.2405	6.34 0.1515 0.1504	0.05 0.0069 0.0066	6.29 0.0187 0.0183	0.2301 0.2209	0.6174 0.6103						
621	7	7 6 9	1.250 0.0012 0.0019	12.77 0.0000 -0.0000	-0.00 -0.0001 -0.0000	0.3744 -0.2547 -0.0760	-0.02 0.4453 -0.0760	6.40 0.1775 0.1773	0.05 0.0076 0.0072	6.32 0.0215 0.0212	0.2161 0.2054	0.6072 0.5991						
621	8	7 6 9	1.251 0.0000 0.0010	0.02 0.0000 -0.0000	-0.00 0.0000 0.0000	-23.6877 -0.1822 -0.1794	-0.01 -5.5023 -0.1794	6.14 0.0454 0.0403	0.04 0.0035 0.0033	6.11 0.0053 0.0046	0.3941 0.4151	0.5888 0.5796						
622	1	7 6 9	1.250 0.0009 0.0011	3.04 0.0000 -0.0001	-0.00 -0.0001 -0.0000	-0.1109 -0.5328 -0.2806	-0.02 0.6593 -0.2806	9.15 0.0296 0.0267	0.05 0.0028 0.0026	9.06 0.0034 0.0030	0.4775 0.4961	0.5745 0.5625						
622	2	7 6 9	1.250 0.0004 0.0013	0.05 0.0001 -0.0000	-0.00 0.0000 -0.0000	2.1458 -0.3370 -0.0103	-0.02 0.1843 -0.0103	9.24 0.0702 0.0654	0.05 0.0047 0.0043	9.15 0.0087 0.0079	0.3352 0.3309	0.6196 0.6071						
622	3	7 6 9	1.251 0.0000 0.0013	1.13 0.0002 -0.0000	-0.00 0.0000 -0.0000	138.9638 44.2417 -0.0261	-0.01 -0.4117 -0.0261	9.27 0.0839 0.0785	0.04 0.0052 0.0047	9.17 0.0104 0.0096	0.3100 0.3034	0.6206 0.6136						
622	4	7 6 9	1.250 0.0001 0.0020	3.24 0.0002 0.0000	-0.00 -0.0000 -0.0000	-6.6268 -0.0128 -0.1257	-0.02 -0.6433 -0.1257	9.31 0.1089 0.1035	0.05 0.0058 0.0055	9.21 0.0135 0.0128	0.2700 0.2655	0.6236 0.6190						

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																						
622	5	7	1.251	6.43	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	9.37	9.14	0.0066	0.0175	0.2367	0.6219
		8	-0.0011	0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	0.1414	0.5189	0.0063	0.0171	0.2276	0.6135
		9	0.0028	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.1400	0.2229	0.0063	0.0171	0.2276	0.6135
622	6	7	1.250	9.58	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	9.43	9.16	0.0074	9.30	0.2210	0.6121
		8	-0.0010	0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	0.1694	0.4574	0.0069	0.0207	0.2046	0.6082
		9	0.0017	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.1694	0.1068	0.0069	0.0206	0.2046	0.6082
622	7	7	1.251	12.80	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	9.46	9.31	0.0079	9.33	0.2045	0.5041
		8	-0.0017	0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	0.1934	0.3119	0.0074	0.0233	0.1940	0.5041
		9	0.0011	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.1929	-0.2106	0.0074	0.0229	0.1940	0.5041
622	8	7	1.251	0.05	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	9.23	9.01	0.0046	9.15	0.3324	0.6134
		8	0.0004	0.0001	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0702	0.3391	0.0043	0.0086	0.3326	0.6057
		9	0.0008	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0652	0.3330	0.0043	0.0078	0.3326	0.6057
623	5	7	1.252	-2.99	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	15.03	14.99	0.0049	14.99	0.3442	0.6336
		8	0.0000	0.0001	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0725	4.5988	0.0048	0.0091	0.3442	0.6336
		9	0.0008	0.0001	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0756	0.6069	0.0048	0.0094	0.3442	0.6336
623	6	7	1.252	0.11	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	15.10	15.01	0.0062	15.05	0.2825	0.6341
		8	0.0018	0.0002	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.1109	0.4845	0.0058	0.0140	0.2825	0.6341
		9	0.0016	0.0001	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.1125	0.3209	0.0058	0.0140	0.2825	0.6341
623	7	7	1.250	1.19	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	15.13	15.01	0.0065	15.07	0.2656	0.6295
		8	0.0014	0.0003	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.1237	0.8848	0.0065	0.0155	0.2656	0.6295
		9	0.0016	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.1242	0.3607	0.0065	0.0154	0.2656	0.6295
623	8	7	1.251	3.30	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	15.16	15.02	0.0069	15.11	0.2375	0.6251
		8	0.0014	0.0002	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.1456	0.6635	0.0064	0.0182	0.2375	0.6251
		9	0.0015	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.1459	0.6693	0.0064	0.0180	0.2375	0.6251
623	9	7	1.251	6.47	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	15.21	15.02	0.0069	15.14	0.2189	0.6163
		8	0.0001	0.0001	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.1734	3.2134	0.0075	0.0213	0.2189	0.6163
		9	0.0029	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.1776	0.4036	0.0069	0.0216	0.2189	0.6163
623	10	7	1.253	9.61	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	15.25	15.02	0.0061	15.18	0.2070	0.6060
		8	0.0000	0.0001	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.1967	9.4723	0.0076	0.0238	0.2070	0.6060
		9	0.0027	0.0001	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.1994	-0.3318	0.0076	0.0238	0.2070	0.6060
623	11	7	1.252	12.85	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	-0.000	15.27	15.02	0.0066	15.19	0.1853	0.5978
		8	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.2178	7.7266	0.0080	0.0260	0.1853	0.5978
		9	0.0007	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.2205	1.3137	0.0073	0.0260	0.1853	0.5978

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																				
623	12	7	1.252	0.10	-0.00	1.0240	-0.02	15.11	0.05	15.05	0.2779	0.6270	0.0014	0.0002	0.0001	1.0240	0.5812	0.1113	0.0061	0.0139	0.2573	0.6188
		6	0.0006	-0.0000	0.0001	-0.1939	1.4888	0.1119	0.0057	0.0138												
624	1	7	0.901	-2.96	-0.00	0.5994	-0.02	15.04	0.05	15.02	0.5211	0.6023	0.0012	0.0001	0.0000	-0.4264	0.5241	0.0808	0.0084	0.0097	0.5059	0.5938
		6	0.0012	0.0001	0.0001	0.5994	0.5886	0.0832	0.0084	0.0098												
		9	0.0008	-0.0000	0.0000	-0.4264	0.5886	0.0832	0.0084	0.0098												
624	2	7	0.900	0.11	-0.00	0.5921	-0.02	15.10	0.05	15.06	0.3972	0.5896	0.0026	0.0003	0.0002	0.5921	0.5224	0.1217	0.0096	0.0143	0.3870	0.5815
		6	0.0026	0.0000	0.0001	0.5921	0.6144	0.1217	0.0094	0.0141												
		9	0.0014	0.0000	0.0001	0.5921	0.6144	0.1217	0.0094	0.0141												
624	3	7	0.902	1.10	-0.00	0.6828	-0.02	15.11	0.05	15.06	0.3626	0.5865	0.0027	0.0003	0.0003	0.6828	0.5635	0.1338	0.0097	0.0157	0.3571	0.5738
		8	0.0027	0.0000	0.0002	0.6828	0.9891	0.1313	0.0093	0.0150												
		9	0.0010	0.0000	0.0002	0.6828	0.9891	0.1313	0.0093	0.0150												
624	4	7	0.900	3.22	-0.00	1.4595	-0.02	15.11	0.05	15.07	0.3118	0.5746	0.0014	0.0004	0.0003	1.4595	1.1655	0.1496	0.0093	0.0172	0.3028	0.5696
		8	0.0014	0.0000	0.0002	1.4595	1.5701	0.1490	0.0090	0.0169												
		9	0.0008	0.0001	0.0002	1.4595	1.5701	0.1490	0.0090	0.0169												
624	5	7	0.900	6.31	-0.00	0.5778	-0.02	15.12	0.06	15.07	0.2508	0.5684	0.0009	0.0001	0.0002	0.5778	0.6820	0.1684	0.0084	0.0191	0.2398	0.5609
		8	0.0009	0.0001	0.0002	0.5778	0.6205	0.1688	0.0081	0.0189												
		9	0.0023	0.0001	0.0002	0.5778	0.6205	0.1688	0.0081	0.0189												
624	6	7	0.900	9.39	-0.00	0.325	-0.02	15.11	0.06	15.05	0.2229	0.5645	0.0008	0.0000	0.0003	0.325	0.3074	0.1755	0.0078	0.0198	0.2095	0.5610
		8	0.0008	0.0000	0.0003	0.325	0.8488	0.1781	0.0074	0.0199												
		9	0.0017	0.0000	0.0003	0.325	0.8488	0.1781	0.0074	0.0199												
624	7	7	0.898	12.56	0.00	-0.0540	-0.02	15.10	0.06	15.05	0.2088	0.5617	-0.0000	0.0000	0.0000	-0.0540	0.2238	0.1761	0.0073	0.0198	0.1981	0.5577
		6	0.0000	0.0001	0.0001	-0.0540	0.2238	0.1775	0.0070	0.0198												
		9	0.0019	-0.0000	0.0000	-0.0540	0.2238	0.1775	0.0070	0.0198												
624	8	7	0.899	0.07	-0.00	0.8148	-0.02	15.10	0.05	15.06	0.3974	0.5884	0.0022	0.0003	0.0001	0.8148	0.6706	0.1214	0.0096	0.0142	0.3856	0.5814
		8	0.0022	0.0000	0.0001	0.8148	0.6738	0.1212	0.0094	0.0141												
		9	0.0012	0.0000	0.0001	0.8148	0.6738	0.1212	0.0094	0.0141												
625	9	7	0.901	-3.03	-0.00	0.1977	-0.02	9.13	0.05	9.07	0.6965	0.5582	0.0014	0.0000	0.0000	0.1977	0.2175	0.0299	0.0041	0.0033	0.6933	0.5646
		8	0.0014	0.0000	0.0000	0.1977	1.2737	0.0305	0.0042	0.0034												
		9	0.0003	-0.0001	0.0000	-1.8515	1.2737	0.0305	0.0042	0.0034												
625	10	7	0.900	0.03	-0.00	0.7212	-0.01	9.23	0.05	9.16	0.5345	0.5992	0.0022	0.0003	0.0002	0.7212	0.5405	0.0717	0.0076	0.0085	0.5312	0.5905
		8	0.0022	0.0000	0.0001	0.7212	0.3885	0.0706	0.0075	0.0083												
		9	0.0016	-0.0000	0.0001	-0.0991	0.3885	0.0706	0.0075	0.0083												

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																			
623	12	7	1.252	0.10	-0.00	1.0240	-0.02	15.11	0.05	15.05	0.2779	0.6270	0.0014	0.0002	0.0001	0.5812	0.1113	0.0061	0.0139	0.2573	0.6108
		8	0.0006	-0.0000	0.0001	-0.1939	1.4488	0.1119	0.0057	0.0138											
624	1	7	0.901	-2.96	-0.00	0.5994	-0.02	15.04	0.05	15.02	0.5211	0.6023	0.0012	0.0001	0.0000	0.5241	0.0808	0.0084	0.0097	0.5059	0.5938
		8	0.0012	0.0001	0.0001	-0.4264	0.5886	0.0832	0.0084	0.0098											
		9	0.0008	-0.0000	0.0000	0.0858	0.6144	0.1217	0.0094	0.0141											
624	2	7	0.900	0.11	-0.00	0.5921	-0.02	15.10	0.05	15.06	0.3972	0.5896	0.0026	0.0003	0.0002	0.5224	0.1217	0.0096	0.0143	0.3870	0.5815
		8	0.0026	0.0000	0.0001	0.0858	0.6144	0.1217	0.0094	0.0141											
		9	0.0014	0.0000	0.0001	0.0858	0.6144	0.1217	0.0094	0.0141											
624	3	7	0.902	1.10	-0.00	0.6828	-0.02	15.11	0.05	15.06	0.3626	0.5865	0.0027	0.0003	0.0002	0.5635	0.1338	0.0097	0.0157	0.3571	0.5738
		8	0.0027	0.0003	0.0003	0.3178	0.9891	0.1313	0.0093	0.0150											
		9	0.0010	0.0000	0.0002	0.3178	0.9891	0.1313	0.0093	0.0150											
624	4	7	0.900	3.22	-0.00	1.4595	-0.02	15.11	0.05	15.07	0.3118	0.5748	0.0014	0.0004	0.0003	1.1655	0.1496	0.0093	0.0172	0.3028	0.5696
		8	0.0014	0.0004	0.0003	0.7338	1.5701	0.1490	0.0090	0.0169											
		9	0.0008	0.0001	0.0002	0.7338	1.5701	0.1490	0.0090	0.0169											
624	5	7	0.900	6.31	-0.00	0.5778	-0.02	15.12	0.06	15.07	0.2508	0.5684	0.0009	0.0001	0.0002	0.6820	0.1684	0.0084	0.0191	0.2398	0.5609
		8	0.0009	0.0001	0.0002	0.3811	0.6205	0.1688	0.0081	0.0189											
		9	0.0023	0.0001	0.0002	0.3811	0.6205	0.1688	0.0081	0.0189											
624	6	7	0.900	9.39	-0.00	0.0325	-0.02	15.11	0.06	15.05	0.2229	0.5645	0.0008	0.0000	0.0003	0.3074	0.1755	0.0078	0.0198	0.2095	0.5610
		8	0.0008	0.0000	0.0003	0.2741	0.8488	0.1781	0.0074	0.0199											
		9	0.0017	0.0000	0.0003	0.2741	0.8488	0.1781	0.0074	0.0199											
624	7	7	0.898	12.56	0.00	-0.0540	-0.02	15.10	0.06	15.05	0.2088	0.5617	-0.0000	0.0000	0.0000	0.2238	0.1761	0.0073	0.0198	0.1981	0.5577
		8	0.0000	0.0001	0.0001	-0.0540	0.2238	0.1775	0.0070	0.0198											
		9	0.0019	-0.0000	0.0000	-0.0540	0.2238	0.1775	0.0070	0.0198											
624	8	7	0.899	0.07	-0.00	0.8148	-0.02	15.10	0.05	15.06	0.3974	0.5884	0.0022	0.0003	0.0001	0.6706	0.1214	0.0096	0.0142	0.3886	0.5814
		8	0.0022	0.0003	0.0001	0.1291	0.6738	0.1212	0.0094	0.0141											
		9	0.0012	0.0000	0.0001	0.1291	0.6738	0.1212	0.0094	0.0141											
625	9	7	0.901	-3.03	-0.00	0.1977	-0.02	9.13	0.05	9.07	0.6965	0.5582	0.0014	0.0000	0.0000	0.2175	0.0299	0.0041	0.0033	0.6933	0.5646
		8	0.0014	0.0000	0.0000	-1.8515	1.2737	0.0305	0.0042	0.0034											
		9	0.0003	-0.0001	0.0000	-1.8515	1.2737	0.0305	0.0042	0.0034											
625	10	7	0.900	0.03	-0.00	0.7212	-0.01	9.23	0.05	9.16	0.5345	0.5992	0.0022	0.0003	0.0001	0.5405	0.0717	0.0076	0.0085	0.5312	0.5905
		8	0.0022	0.0003	0.0002	0.0991	0.3885	0.0706	0.0075	0.0083											
		9	0.0016	-0.0000	0.0001	0.0991	0.3885	0.0706	0.0075	0.0083											

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd																				
625	11	7	0.900	1.08	-0.00	1.2995	-0.01	9.26	0.06	9.18	0.4972	0.6003	0.0018	0.0004	0.0003	0.8393	0.0863	0.085	0.0103	0.4974	0.5910
		8	0.0021	-0.0000	0.0000	-0.0675	0.2219	0.0839	0.083	0.0099											
		9																			
625	12	7	0.899	3.19	-0.00	2.6102	-0.02	9.30	0.05	9.22	0.4176	0.5924	0.0008	0.0004	0.0002	1.3176	0.1141	0.095	0.0135	0.4168	0.5852
		8	0.0021	0.0000	0.0001	0.0305	0.3036	0.1118	0.093	0.0130											
		9																			
625	13	7	0.900	6.30	-0.00	0.5915	-0.02	9.34	0.06	9.24	0.3273	0.5857	0.0006	0.0000	0.0000	0.5883	0.1489	0.097	0.0174	0.3157	0.5769
		8	0.0022	0.0001	0.0002	0.3817	0.6428	0.1492	0.094	0.0172											
		9																			
625	14	7	0.901	9.39	-0.00	0.6598	-0.02	9.34	0.05	9.24	0.2592	0.5693	0.0008	0.0001	0.0002	0.388	0.1678	0.087	0.0191	0.2466	0.5619
		8	0.0026	0.0000	0.0002	0.1557	0.4674	0.1679	0.082	0.0168											
		9																			
625	15	7	0.900	12.56	0.00	0.8628	-0.02	9.34	0.06	9.22	0.2199	0.5666	0.0003	0.0000	0.0000	0.9879	0.1787	0.078	0.0202	0.2052	0.5598
		8	0.0018	0.0000	0.0001	0.0757	0.3562	0.1803	0.074	0.0201											
		9																			
625	16	7	0.900	0.01	-0.00	0.4322	-0.02	9.23	0.06	9.15	0.5327	0.5955	0.0028	0.0002	0.0001	0.3280	0.0717	0.076	0.0085	0.5311	0.5867
		8	0.0009	0.0000	0.0001	0.2107	1.0109	0.0706	0.075	0.0082											
		9																			
626	1	7	0.900	-3.05	-0.00	1.2694	-0.02	6.04	0.05	6.02	0.8669	0.5001	0.0007	0.0001	0.0001	0.9847	0.0096	0.016	0.0009	1.0468	0.4921
		8	0.0014	-0.0001	-0.0000	-0.6558	-0.1122	0.0086	0.018	0.0008											
		9																			
626	2	7	0.900	-0.00	-0.00	0.5970	-0.02	6.13	0.05	6.10	0.6126	0.5785	0.0025	0.0003	0.0002	0.5309	0.0446	0.054	0.0051	0.6133	0.5664
		8	0.0015	-0.0000	0.0001	-0.1165	0.3984	0.0434	0.053	0.0049											
		9																			
626	3	7	0.901	1.06	-0.00	0.8615	-0.02	6.17	0.06	6.12	0.5517	0.5905	0.0023	0.0004	0.0003	0.6481	0.0600	0.066	0.0070	0.5605	0.5832
		8	0.0018	-0.0000	0.0001	-0.0430	0.3817	0.0571	0.064	0.0066											
		9																			
626	4	7	0.900	3.15	-0.00	1.1245	-0.02	6.23	0.06	6.18	0.4891	0.5970	0.0015	0.0003	0.0001	0.4843	0.0882	0.086	0.0105	0.4911	0.5896
		8	0.0018	0.0000	0.0001	0.0823	0.4602	0.0856	0.084	0.0101											
		9																			
626	5	7	0.901	6.26	-0.00	1.6273	-0.02	6.28	0.06	6.23	0.3852	0.5930	0.0004	0.0001	0.0000	0.7416	0.1275	0.098	0.0151	0.3734	0.5777
		8	0.0028	0.0001	0.0002	0.2724	0.4597	0.1287	0.096	0.0148											
		9																			

See Key

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key														
626	6	0.950	9.38	-0.00	5.5502	-0.02	6.31	0.06	6.24	0.3068	0.5774					
	8	0.0001	0.0000	0.0001	0.0217	5.1842	0.1566	0.0096	0.0180	0.2886	0.5652					
	9	0.0034	0.0000	0.0001	0.0217	0.2556	0.1575	0.0090	0.0178							
626	7	0.900	12.56	-0.00	0.2202	-0.02	6.29	0.06	6.23	0.2351	0.5657					
	8	0.0005	0.0000	0.0000	0.0388	0.0794	0.1711	0.0080	0.0193	0.2243	0.5596					
	9	0.0025	-0.0000	0.0000	-0.0388	0.1265	0.1741	0.0078	0.0194							
626	8	0.900	0.00	-0.00	0.6569	-0.02	6.13	0.06	6.10	0.6076	0.5715					
	8	0.0024	0.0003	0.0002	0.0683	0.5566	0.0449	0.0054	0.0051	0.6090	0.5615					
	9	0.0012	0.0000	0.0001	0.0683	0.7060	0.0437	0.0053	0.0049							
627	1	0.900	-3.06	-0.00	0.3468	-0.02	2.94	0.05	3.02	0.6556	0.6979					
	8	0.0014	0.0001	0.0001	0.3468	0.3632	-0.0054	-0.0009	-0.0007	0.4354	0.6654					
	9	0.0003	-0.0001	0.0000	-1.9943	0.9715	-0.0068	-0.0006	-0.0009							
627	2	0.899	-0.02	-0.00	0.8321	-0.02	3.04	0.05	3.11	0.6476	0.5537					
	8	0.0021	0.0003	0.0002	0.8321	0.6424	0.222	0.0028	0.0024	0.7037	0.5598					
	9	0.0016	-0.0000	0.0001	-0.1646	0.3361	0.0202	0.0028	0.0021							
627	3	0.900	1.01	-0.00	0.8294	-0.01	3.08	0.06	3.14	0.6256	0.5586					
	8	0.0023	0.0003	0.0002	0.8294	0.6207	0.340	0.0042	0.0038	0.6539	0.5567					
	9	0.0019	-0.0000	0.0001	-0.1133	0.2892	0.0309	0.0040	0.0034							
627	4	0.900	3.11	-0.00	1.7313	-0.02	3.14	0.05	3.20	0.5382	0.5903					
	8	0.0011	0.0003	0.0002	1.7313	0.9518	0.0612	0.0065	0.0072	0.5520	0.5803					
	9	0.0019	0.0000	0.0001	0.0595	0.4202	0.0580	0.0064	0.0067							
627	5	0.901	6.25	-0.00	4.4791	-0.02	3.22	0.06	3.28	0.4490	0.5941					
	8	0.0002	0.0002	0.0000	4.4791	1.7076	0.1024	0.0092	0.0121	0.4375	0.5831					
	9	0.0032	0.0001	0.0002	0.2102	0.3710	0.1041	0.0091	0.0121							
627	6	0.899	9.35	-0.00	0.3236	-0.02	3.28	0.06	3.32	0.3485	0.5925					
	8	0.0008	0.0000	0.0000	0.1516	0.3698	0.1441	0.0100	0.0170	0.3343	0.5807					
	9	0.0027	0.0000	0.0002	0.1516	0.4701	0.1451	0.0097	0.0168							
627	7	0.899	12.54	0.00	1.2216	-0.02	3.21	0.06	3.32	0.2710	0.5703					
	8	0.0003	0.0000	0.0001	1.2216	1.3312	0.1651	0.0089	0.0188	0.2559	0.5608					
	9	0.0022	0.0000	0.0001	0.0257	0.2353	0.1661	0.0085	0.0186							
627	8	0.901	-0.02	-0.00	0.6253	-0.02	3.04	0.06	3.11	0.6370	0.5367					
	8	0.0023	0.0002	0.0002	0.6253	0.4881	0.224	0.0028	0.0024	0.6920	0.5317					
	9	0.0016	-0.0000	0.0000	-0.1195	0.3015	0.0205	0.0028	0.0021							

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key															
628	1	7	0.899	-3.09	-0.00	0.2860	-0.2340	0.930	-0.0190	0.05	-0.0025	0.088	0.6762	0.5973			
	8	9	0.0013	0.0000	0.0000	-1.6126	0.4506	-0.0203	-0.0203	-0.0023	-0.0024	0.5754	0.6068				
	7	8	0.898	-0.04	-0.00	1.0512	-0.02	1.03	0.089	0.05	0.0009	0.6685	0.5286				
62A	2	9	0.0018	0.0003	0.0002	-0.2264	0.7926	0.0059	0.0012	0.0010	0.0006	0.9129	0.5364				
	7	8	0.900	1.01	-0.00	0.6651	-0.4863	1.06	0.0158	0.05	0.0021	0.6112	0.5490				
628	3	9	0.0025	0.0003	0.0002	-0.1208	0.6122	0.0158	0.0158	0.0022	0.0016	0.7120	0.5323				
	7	8	0.900	3.10	-0.00	0.9723	-0.01	1.13	0.0439	0.05	1.05	0.5775	0.5668				
628	4	9	0.0016	0.0003	0.0002	-0.0567	0.6446	0.0392	0.0050	0.0048	0.0049	0.6109	0.5610				
	7	8	0.901	6.23	-0.00	3.1751	-0.5333	1.22	0.0844	0.06	1.15	0.4885	0.5927				
62A	5	9	0.0029	0.0001	0.0002	0.2352	0.3969	0.0847	0.0847	0.0082	0.0099	0.4879	0.5878				
	7	8	0.901	9.34	-0.00	0.9649	-0.3083	1.28	0.1264	0.05	1.19	0.3912	0.5933				
628	6	9	0.0022	0.0001	0.0003	0.2587	0.6778	0.1273	0.0098	0.0096	0.0148	0.3795	0.5823				
	7	8	0.898	12.53	0.00	-13.6750	-0.02	1.30	0.1573	0.06	1.21	0.3046	0.5733				
62A	7	9	-0.0017	0.0000	0.0001	0.0665	-0.3814	0.1573	0.0091	0.0091	0.0177	0.2905	0.5654				
	7	8	0.901	-0.04	-0.00	1.1541	-0.02	1.02	0.0058	0.05	0.95	0.6446	0.4996				
629	8	9	0.0014	0.0000	0.0003	-0.1224	0.3405	0.0058	0.0010	0.0010	0.0005	0.9124	0.5064				
	7	8	0.900	-3.10	-0.00	0.1565	-0.02	-0.15	-0.0035	0.05	-0.09	0.6507	0.5845				
629	1	9	0.0014	0.0000	0.0000	-2.2372	0.4577	-0.0270	-0.0031	-0.0032	-0.0032	0.5857	0.5910				
	7	8	0.898	-0.06	-0.00	0.8226	-0.01	-0.06	0.0002	0.05	-0.0023	0.4136	0.4822				
629	2	9	0.0021	0.0003	0.0002	-0.1046	0.6463	0.0005	0.0005	0.0002	-0.0000	2.3473	-0.1855				
	7	8	0.900	1.01	-0.00	0.7145	-0.02	-0.03	0.0016	0.06	0.0003	0.6315	0.5232				
629	3	9	0.0026	0.0003	0.0002	-0.0944	0.4547	0.0090	0.0014	0.0014	0.0009	0.7938	0.5025				
	7	8	0.901	-0.00	-0.00	0.0944	-0.4547	0.0090	0.0014	0.0014	0.0009	0.7938	0.5025				

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key															
629	4	7	0.900	5.07	-0.00	1.1535	-0.02	0.6881	0.0350	0.06	0.05	0.5884	0.5553					
		8	0.0015	0.0003	0.0002	0.0202	0.3233	0.0315	0.0041	0.0038		0.6185	0.5615					
		9	0.0019	0.0000	0.0001	0.0202	0.3233	0.0315	0.0039	0.0035								
629	5	7	0.89A	6.22	-0.00	0.8646	-0.02	0.5334	0.0750	0.06	0.15	0.5007	0.5935					
		8	0.0006	0.0001	0.0000	0.1459	0.2327	0.0750	0.0075	0.0089		0.5038	0.5658					
		9	0.0035	0.0001	0.0001	0.1459	0.2327	0.0750	0.0076	0.0089								
629	6	7	0.897	9.35	-0.00	2.3576	-0.02	0.6267	0.1173	0.05	0.21	0.4139	0.5924					
		8	0.0002	0.0000	0.0000	0.0936	0.3449	0.1186	0.0097	0.0139		0.4041	0.5652					
		9	0.0031	0.0000	0.0002	0.0936	0.3449	0.1186	0.0095	0.0138								
629	7	7	0.897	12.52	0.00	-6.9870	-0.02	0.2795	0.1522	0.06	0.24	0.3210	0.5794					
		8	0.0000	0.0001	0.0000	0.0016	0.1A13	0.1522	0.0097	0.0176		0.3088	0.5709					
		9	-0.0021	0.0000	0.0000	0.0016	0.1A13	0.1522	0.0094	0.0173								
629	8	7	0.898	-0.05	-0.00	0.7976	-0.02	0.5661	0.0034	0.06	0.03	0.3423	0.4281					
		8	0.0021	0.0003	0.0002	0.7976	0.5661	0.0034	0.0002	0.0002		1.8140	-0.2109					
		9	0.0008	0.0000	0.0001	0.0605	0.8152	0.0007	0.0002	-0.0000								
630	1	7	0.89A	-3.11	-0.00	0.2257	-0.02	0.0360	-0.0511	0.05	0.08	0.5827	0.5843					
		8	0.0011	0.0000	-0.0000	2.3562	-1.1545	-0.0505	-0.0059	-0.0059		0.5656	0.5849					
		9	-0.0001	-0.0000	0.0000	2.3562	-1.1545	-0.0505	-0.0057	-0.0059								
630	2	7	0.898	-0.07	-0.00	1.1563	0.7712	-0.0130	0.05	0.05	0.00	0.8429	0.5685					
		8	0.0018	0.0004	0.0002	1.1563	0.7712	-0.0130	-0.0022	-0.0014		0.7187	0.5688					
		9	0.0017	-0.0000	0.0000	-0.1540	0.1888	-0.0148	-0.0021	-0.0017								
630	3	7	0.897	0.99	-0.00	1.3623	0.8792	-0.0294	0.05	0.05	0.97	1.7947	0.5996					
		8	0.0017	0.0004	0.0003	1.3623	0.8792	-0.0294	-0.0008	-0.0002		0.8912	0.5867					
		9	0.0016	-0.0000	0.0000	-0.1392	0.2539	-0.0053	-0.0009	-0.0006								
630	4	7	0.89A	3.05	-0.00	0.7799	0.3932	-0.0145	0.06	0.06	0.92	0.5629	0.5485					
		8	0.0021	0.0003	0.0001	0.7799	0.3932	0.0145	0.0016	0.0015		0.6501	0.5496					
		9	0.001A	0.0000	0.0001	0.0472	0.4538	0.0107	0.0013	0.0011								
630	5	7	0.899	6.17	-0.00	0.2502	0.3211	-0.0479	0.06	0.06	0.81	0.5463	0.5690					
		8	0.0011	0.0000	0.0000	0.2502	0.3211	-0.0479	0.0052	0.0054		0.5663	0.5598					
		9	0.0031	0.0001	0.0002	0.2159	0.3648	0.0483	0.0054	0.0054								
630	6	7	0.899	9.28	-0.00	0.4756	-0.02	0.2960	0.06	0.06	0.72	0.4744	0.5917					
		8	0.0005	0.0000	0.0000	0.4756	0.2960	0.0902	0.0085	0.0106		0.4637	0.5846					
		9	0.0024	0.0001	0.0003	0.2671	0.7010	0.0924	0.0085	0.0108								

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key															
630	7	7	0.899	12.50	0.000	-0.000	0.000	-0.0278	-0.02	-2.63	0.06	-2.68	0.3688	0.5940				
		6	0.0007	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.3663	0.1355	0.0099	0.0161	0.3564	0.5836				
		5	0.0014	0.0000	0.0001	0.0001	0.0001	0.1829	0.6120	0.1365	0.0097	0.0159						
630	8	7	0.899	-0.006	-0.000	-0.000	0.4309	-0.02	-0.02	-2.98	0.05	-2.99	0.8846	0.6057				
		6	0.0028	0.0002	0.0001	0.0001	0.2965	0.3116	-0.0124	-0.0142	-0.0021	-0.0015	0.7211	0.6155				
		5	0.0004	0.0000	0.0001	0.0001	0.2965	1.6576	-0.0142	-0.0142	-0.0020	-0.0017						
632	3	7	1.252	-3.16	-0.000	-0.000	8.7485	-0.03	-3.01	0.04	-3.02	0.1024	0.6769					
		6	-0.0001	-0.0003	-0.0000	-0.0000	0.0098	0.4376	-0.0445	-0.0009	-0.0060	0.1381	0.6751					
		5	-0.0046	-0.0000	-0.0007	-0.0007	0.0098	0.8162	-0.0435	-0.0012	-0.0058							
632	4	7	1.251	-1.11	-0.000	-0.000	-3.2514	-0.03	-2.98	0.05	-3.00	0.1105	0.6882					
		6	0.0004	-0.0002	0.0001	0.0001	-0.0128	1.6882	-0.0233	-0.0008	-0.0032	0.1747	0.6959					
		5	-0.0037	0.0000	-0.0007	-0.0007	-0.0128	1.0224	-0.0237	-0.0008	-0.0032							
632	5	7	1.255	-0.005	-0.000	-0.000	-2.2145	-0.03	-2.96	0.05	-2.98	0.1199	0.7203					
		6	0.0005	-0.0002	0.0002	0.0002	-0.0284	1.8961	-0.0133	-0.0003	-0.0019	0.2298	0.7127					
		5	-0.0041	0.0000	-0.0007	-0.0007	-0.0284	0.8827	-0.0144	-0.0006	-0.0020							
632	6	7	1.255	0.47	-0.000	-0.000	-1.0191	-0.02	-2.95	0.05	-2.98	0.1162	0.7564					
		6	0.0016	-0.0003	0.0001	0.0001	-0.0604	0.4908	-0.0086	-0.0002	-0.0013	0.2896	0.7107					
		5	-0.0041	0.0000	-0.0006	-0.0006	-0.0604	0.7821	-0.0101	-0.0005	-0.0014							
632	7	7	1.252	0.99	-0.000	-0.000	-0.9293	-0.02	-2.94	0.05	-2.97	0.1041	0.8066					
		6	0.0014	-0.0000	0.0001	0.0001	-0.0001	0.6900	-0.0040	-0.0005	-0.0008	0.4778	0.8066					
		5	-0.0036	0.0000	-0.0007	-0.0007	-0.0001	0.9866	-0.0053	-0.0005	-0.0008							
632	8	7	1.252	3.12	-0.000	-0.000	-0.9976	-0.02	-2.91	0.05	-2.95	0.1475	0.8324					
		6	0.0012	-0.0002	0.0001	0.0001	-0.0395	0.5596	0.0127	0.0003	0.0016	0.0062	0.8592					
		5	-0.0035	0.0000	-0.0005	-0.0005	-0.0395	0.8285	0.0096	0.0000	0.0012							
632	9	7	1.252	6.24	-0.000	-0.000	1.1768	-0.03	-2.86	0.05	-2.90	0.1234	0.8561					
		6	0.0008	-0.0002	0.0000	0.0000	-0.1479	0.2615	0.0398	0.0007	0.0051	0.0997	0.8491					
		5	-0.0025	0.0000	-0.0004	-0.0004	-0.1479	0.9229	0.0398	0.0007	0.0051							
632	10	7	1.253	9.39	-0.000	-0.000	2.0945	-0.03	-2.82	0.04	-2.87	0.1115	0.8481					
		6	0.0005	-0.0002	0.0000	0.0000	0.0185	0.6808	0.0669	0.0014	0.0085	0.0999	0.8481					
		5	-0.0046	-0.0000	-0.0007	-0.0007	0.0185	0.7843	0.0662	0.0013	0.0085							
632	11	7	1.253	12.61	-0.000	-0.000	0.8407	-0.03	-2.79	0.04	-2.84	0.1029	0.8285					
		6	0.0011	-0.0001	0.0001	0.0001	-0.0078	0.7152	0.0889	0.0018	0.0112	0.0945	0.8285					
		5	-0.0048	-0.0000	-0.0007	-0.0007	-0.0078	0.7441	0.0886	0.0016	0.0111							

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																
632	12	7	1.251	-0.006	-0.000	-0.001	-1.218	0.730	-2.96	0.003	-0.003	0.05	-2.99	0.1397	0.7460				
		6	0.0012	-0.0003	0.0001	-0.0007	-0.0860	0.7306	-0.0132	-0.0003	-0.0006	-0.0019	0.2322	0.7248					
		9	-0.0042	0.0000	-0.0007	-0.0007	-0.0860	0.8343	-0.0149	-0.0006	-0.0021	-0.0021	0.2322	0.7248					
633	1	7	1.251	-3.12	-0.000	-0.000	3.1008	-0.03	-1.00	0.005	0.05	-1.05	0.1108	0.6821					
		6	-0.0004	-0.0002	-0.0000	-0.0008	-0.0414	0.1977	-0.0319	-0.0007	-0.0043	-0.0043	0.1500	0.6763					
		9	-0.0046	0.0000	-0.0008	-0.0008	-0.0414	0.8860	-0.0319	-0.0009	-0.0043	-0.0043	0.1500	0.6763					
633	2	7	1.252	-1.08	-0.000	-0.000	-5.3005	-0.03	-0.96	0.005	0.05	-1.02	0.1382	0.7147					
		6	0.0002	-0.0002	0.0001	-0.0007	-0.0539	2.3481	-0.0110	-0.0003	-0.0015	-0.0015	0.2551	0.6943					
		9	-0.0038	0.0000	-0.0007	-0.0007	-0.0539	1.0271	-0.0123	-0.0006	-0.0017	-0.0017	0.2551	0.6943					
633	3	7	1.255	-0.04	-0.000	-0.000	-1.9841	-0.03	-0.95	0.005	0.05	-1.01	0.1094	0.8322					
		6	0.0006	-0.0002	0.0001	-0.0007	-0.0433	1.4032	-0.0027	-0.0004	-0.0005	-0.0005	0.5276	0.7059					
		9	-0.0040	0.0000	-0.0007	-0.0007	-0.0433	0.9385	-0.0040	-0.0004	-0.0005	-0.0005	0.5276	0.7059					
633	4	7	1.254	0.48	-0.000	-0.000	-1.5433	-0.03	-0.94	0.005	0.05	-1.00	0.6343	0.6325					
		6	0.0009	-0.0002	0.0001	-0.0007	-0.0223	0.7701	0.0009	0.0001	0.0001	0.0001	2.2629	0.5277					
		9	-0.0036	0.0000	-0.0007	-0.0007	-0.0223	1.0637	-0.0006	-0.0003	-0.0000	-0.0000	2.2629	0.5277					
633	5	7	1.252	1.00	-0.000	-0.000	-1.2924	-0.02	-0.93	0.005	0.05	-1.00	0.1965	0.6414					
		6	0.000A	-0.0002	0.0001	-0.0007	-0.0317	1.1606	0.0059	0.0002	0.0002	0.0002	0.2468	0.7357					
		9	-0.0037	0.0000	-0.0007	-0.0007	-0.0317	0.9979	0.0030	-0.0001	-0.0001	-0.0001	0.2468	0.7357					
633	6	7	1.250	3.13	-0.000	-0.000	-1.0839	-0.02	-0.90	0.006	0.05	-0.97	0.1192	0.6480					
		6	0.0011	-0.0002	0.0000	-0.0006	-0.0527	0.3362	0.0252	0.0006	0.0003	0.0003	0.0716	0.6497					
		9	-0.0035	0.0000	-0.0006	-0.0006	-0.0527	0.9274	0.0209	0.0003	0.0003	0.0003	0.0716	0.6497					
633	7	7	1.251	6.26	-0.000	-0.000	1.7799	-0.03	-0.85	0.005	0.05	-0.92	0.1119	0.6599					
		6	0.0007	-0.0002	0.0000	-0.0004	-0.1795	0.5488	0.0523	0.0011	0.0009	0.0009	0.0956	0.6511					
		9	-0.0030	0.0001	-0.0004	-0.0004	-0.1795	0.7847	0.0515	0.0009	0.0009	0.0009	0.0956	0.6511					
633	8	7	1.251	9.42	-0.000	-0.000	1.2329	-0.02	-0.81	0.004	0.04	-0.88	0.1049	0.6445					
		6	0.0008	-0.0002	0.0000	-0.0007	0.0190	0.5269	0.0775	0.0016	0.0014	0.0014	0.0941	0.6344					
		9	-0.0047	-0.0000	-0.0007	-0.0007	0.0190	0.8172	0.0765	0.0014	0.0014	0.0014	0.0941	0.6344					
633	9	7	1.253	12.64	-0.000	-0.000	0.5022	-0.03	-0.78	0.001	0.04	-0.86	0.0962	0.6264					
		6	0.0016	-0.0001	-0.0001	-0.0008	0.0645	0.5772	0.0985	0.0018	0.0017	0.0017	0.0895	0.6211					
		9	-0.0041	-0.0000	-0.0008	-0.0008	0.0645	0.9799	0.0978	0.0017	0.0017	0.0017	0.0895	0.6211					
633	10	7	1.252	-0.06	-0.000	-0.000	-1.5722	-0.03	-0.94	0.005	0.05	-1.01	0.2145	1.0385					
		6	0.0009	-0.0003	0.0001	-0.0007	-0.1045	0.6913	-0.0024	-0.0001	-0.0005	-0.0005	0.5116	0.7715					
		9	-0.0044	0.0000	-0.0007	-0.0007	-0.1045	0.8310	-0.0042	-0.0004	-0.0006	-0.0006	0.5116	0.7715					

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																								
634	1	7	1.252	-3.13	-0.000	5.0748	-0.03	-0.11	0.05	-0.07	0.1164	0.6864	7	6	9	1.252	-0.0003	-0.0000	-0.0008	0.9122	-0.0262	-0.0006	-0.0036	0.1694	0.6890	
			-0.00046	0.0000	-0.0008	-0.0601	0.8765	-0.0257	-0.0008	-0.0035																
634	2	7	1.251	-1.05	-0.000	-3.1476	-0.03	-0.08	0.05	-0.04	0.1392	0.7545	7	6	9	1.251	-0.0004	-0.0008	-0.0008	0.8296	-0.0059	-0.0001	-0.0008	0.3815	0.7199	
			-0.00037	0.0000	-0.0008	-0.0543	1.0778	-0.0067	-0.0005	-0.0009																
634	3	7	1.251	-0.03	-0.000	-1.8757	-0.02	-0.06	0.05	-0.02	0.3972	0.6654	7	6	9	1.251	-0.0007	-0.0001	-0.0007	0.7706	0.0015	0.0001	0.0002	0.3668	0.6812	
			0.0007	-0.0002	0.0001	-0.0671	0.8954	0.0003	-0.0002	0.0000																
634	4	7	1.253	0.52	-0.000	-1.6675	-0.02	-0.06	0.05	-0.02	0.1896	0.6385	7	6	9	1.253	-0.0008	-0.0001	-0.0007	0.8043	0.0062	0.0002	0.0008	0.1326	0.6292	
			-0.00039	0.0000	-0.0007	-0.0422	0.9808	0.0040	-0.0001	0.0005																
634	5	7	1.253	1.00	-0.000	-1.6119	-0.03	-0.05	0.05	-0.02	0.1659	0.6577	7	6	9	1.253	-0.0007	-0.0001	-0.0007	1.0224	0.0107	0.0003	0.0014	0.0029	0.6678	
			0.0007	-0.0002	0.0001	-0.0477	1.0039	0.0080	0.0000	0.0010																
634	6	7	1.253	3.15	-0.000	-2.1210	-0.02	-0.01	0.05	-0.00	0.1124	0.6581	7	6	9	1.253	-0.0004	-0.0001	-0.0006	1.4679	0.0309	0.0006	0.0040	0.0816	0.6487	
			-0.00033	0.0000	-0.0006	-0.0444	1.0052	0.0272	0.0004	0.0035																
634	7	7	1.251	6.27	-0.000	1.4994	-0.03	0.04	0.05	0.04	0.1085	0.6579	7	6	9	1.251	-0.0008	-0.0002	-0.0005	0.4981	0.0575	0.0012	0.0074	0.0964	0.6540	
			-0.00023	0.0000	-0.0005	-0.1781	1.1537	0.0571	0.0011	0.0074																
634	8	7	1.253	9.42	-0.000	1.4238	-0.03	0.06	0.04	0.07	0.1016	0.6389	7	6	9	1.253	-0.0007	-0.0000	-0.0007	0.6036	0.0822	0.0015	0.0105	0.0937	0.6363	
			-0.00046	-0.0002	-0.0007	0.0047	0.8339	0.0812	0.0015	0.0103																
634	9	7	1.254	12.64	-0.000	0.7261	-0.03	0.10	0.05	0.10	0.0930	0.6230	7	6	9	1.254	-0.0013	-0.0001	-0.0008	0.7545	0.1025	0.0019	0.0127	0.0877	0.6167	
			-0.00040	-0.0000	-0.0008	0.0695	1.0195	0.1019	0.0017	0.0125																
634	10	7	1.252	-0.05	-0.000	-1.9225	-0.03	-0.06	0.05	-0.03	0.2665	0.4684	7	6	9	1.252	-0.0008	-0.0003	-0.0007	0.6083	0.0015	0.0000	0.0001	0.2665	0.4684	
			0.00043	0.0000	-0.0007	-0.0914	0.8612	0.0001	-0.0002	-0.0000																
635	8	7	1.250	-3.13	-0.000	1.2891	-0.03	0.96	0.04	0.92	0.0959	0.6575	7	6	9	1.250	-0.0009	-0.0002	-0.0008	0.1879	-0.0202	-0.0007	-0.0026	0.1883	0.6612	
			-0.00045	-0.0000	-0.0008	0.0419	0.9055	-0.0194	-0.0007	-0.0025																

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																		
635	9	1.251	-1.06	-0.00	-7.7482	-0.03	1.00	0.05	0.94	-0.5443	0.6154	0.0001	0.0000	0.0001	0.0000	0.0000	0.0000	0.0000	0.0001	0.4799
	8	-0.0040	-0.0000	0.0007	0.0144	3.4689	-0.0007	0.0000	-0.0001	-1.5691	0.0001	-0.0001	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0003	-0.0001	0.0001
	9	-0.0040	-0.0000	-0.0007	0.0144	0.9499	-0.0010	-0.0010	0.0007	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0001	0.0001
635	10	1.250	-0.03	-0.00	-3.2898	-0.03	1.01	0.05	0.95	0.2513	0.7000	0.0003	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0009	0.7617
	8	0.0003	-0.0002	0.0002	0.0120	2.7137	0.0069	0.0003	0.0009	0.0000	0.0000	0.0000	0.0003	0.0003	0.0003	0.0003	0.0003	0.0003	0.0007	0.0007
	9	-0.0040	-0.0000	-0.0007	0.0120	0.9641	0.0048	-0.0000	0.0048	-0.0260	0.0000	0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	0.0007	0.0007
635	11	1.251	0.51	-0.00	-1.8532	-0.03	1.03	0.05	0.96	0.1906	0.6742	0.0007	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0015	0.7243
	8	0.0007	-0.0002	0.0001	-0.0044	1.2294	0.0116	0.0004	0.0013	0.0000	0.0000	0.0000	0.0004	0.0004	0.0004	0.0004	0.0004	0.0004	0.0015	0.0015
	9	-0.0041	0.0000	-0.0007	-0.0044	0.8823	0.0091	0.0000	0.0091	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0013	0.0013
635	12	1.250	1.00	-0.00	-1.5017	-0.03	1.03	0.05	0.97	0.1605	0.6625	0.0008	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0022	0.6808
	8	0.0008	-0.0002	0.0002	0.0016	1.2170	0.0167	0.0005	0.0167	0.0000	0.0000	0.0000	0.0005	0.0005	0.0005	0.0005	0.0005	0.0005	0.0022	0.0022
	9	-0.0039	-0.0000	-0.0007	0.0016	0.9110	0.0138	0.0001	0.0138	0.0000	0.0000	0.0000	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0018	0.0018
635	13	1.251	3.14	-0.00	-2.4028	-0.03	1.07	0.04	1.00	0.1204	0.6704	0.0001	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	1.00	0.6704
	8	0.0003	-0.0001	0.0001	0.0241	1.9294	0.0371	0.0008	0.0371	0.0000	0.0000	0.0000	0.0008	0.0008	0.0008	0.0008	0.0008	0.0008	1.00	0.0049
	9	-0.0034	-0.0000	-0.0006	-0.0241	0.9127	0.0331	0.0006	0.0331	0.0000	0.0000	0.0000	0.0006	0.0006	0.0006	0.0006	0.0006	0.0006	0.0044	0.6702
635	14	1.251	6.28	-0.00	-1.0126	-0.03	1.11	0.05	1.05	0.1130	0.6561	0.0004	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	1.05	0.6561
	8	0.0011	-0.0002	-0.0000	0.0126	0.1269	0.0631	0.0014	0.0631	0.0000	0.0000	0.0000	0.0014	0.0014	0.0014	0.0014	0.0014	0.0014	1.05	0.0082
	9	-0.0031	0.0001	-0.0004	-0.0126	0.7072	0.0629	0.0012	0.0629	0.0000	0.0000	0.0000	0.0012	0.0012	0.0012	0.0012	0.0012	0.0012	0.0082	0.0082
635	15	1.251	9.43	-0.00	0.6622	-0.03	1.15	0.04	1.07	0.1033	0.6370	0.0007	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	1.07	0.6370
	8	0.0012	-0.0000	-0.0000	0.0756	0.1469	0.0871	0.0016	0.0871	0.0000	0.0000	0.0000	0.0016	0.0016	0.0016	0.0016	0.0016	0.0016	1.07	0.0109
	9	-0.0044	-0.0000	-0.0007	0.0756	0.8633	0.0862	0.0000	0.0862	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0109	0.0109
635	16	1.251	12.65	-0.00	-0.4936	-0.03	1.18	0.04	1.09	0.0943	0.6209	0.0001	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	1.09	0.6209
	8	0.0016	-0.0001	-0.0001	0.4936	0.4611	1.066	0.0020	1.066	0.0000	0.0000	0.0000	0.0020	0.0020	0.0020	0.0020	0.0020	0.0020	1.09	0.0130
	9	-0.0049	-0.0000	-0.0007	0.0557	0.7154	1.061	0.0018	1.061	0.0000	0.0000	0.0000	0.0018	0.0018	0.0018	0.0018	0.0018	0.0018	0.0130	0.0130
635	17	1.252	-0.03	-0.00	-1.9698	-0.03	1.02	0.05	0.95	0.2448	0.7405	0.0007	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.95	0.7405
	8	0.0007	-0.0002	0.0001	0.9698	1.2202	0.067	0.0003	0.067	0.0000	0.0000	0.0000	0.0003	0.0003	0.0003	0.0003	0.0003	0.0003	0.95	0.0006
	9	-0.0041	0.0000	-0.0007	-0.0283	0.9261	0.0045	-0.0000	0.0045	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	0.0006	0.0006
636	1	1.251	-3.11	-0.00	2.9367	-0.03	2.96	0.04	3.02	0.0624	0.6874	0.0003	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	3.02	0.6874
	8	0.0005	-0.0003	-0.0000	2.9367	0.0756	0.085	-0.0001	0.085	0.0000	0.0000	0.0000	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	3.02	0.0011
	9	-0.0050	0.0000	-0.0007	-0.0334	0.7849	-0.0073	-0.0004	-0.0073	-0.0000	-0.0000	-0.0000	-0.0004	-0.0004	-0.0004	-0.0004	-0.0004	-0.0004	0.0011	0.0011
636	2	1.252	-1.03	-0.00	46.3307-21	-0.03	2.99	0.05	3.05	0.2125	0.7104	0.0001	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	3.05	0.7104
	8	0.0000	-0.0002	0.0001	3307-21	0.7084	0.0078	0.0003	0.0078	0.0000	0.0000	0.0000	0.0003	0.0003	0.0003	0.0003	0.0003	0.0003	3.05	0.0012
	9	-0.0043	0.0000	-0.0007	-0.0250	0.8683	-0.0078	0.0000	-0.0078	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0012	0.0012

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																			
636	3	1.251	-0.001	-0.001	-2.2025	-0.03	3.01	0.05	3.06	0.1568	0.6919	0.0006	-0.0002	0.0001	-2.2025	-0.03	3.01	0.0005	0.0024	0.0688	0.6909
	6	0.0040	0.0000	0.0000	0.0084	0.9458	0.0163	0.0002	0.0022												
	9	-0.0040	0.0000	0.0000	-0.0084	0.9458	0.0163	0.0002	0.0022												
636	4	1.252	0.54	-0.001	-2.0207	-0.03	3.02	0.05	3.07	0.1375	0.6811	0.0006	-0.0002	0.0001	-2.0207	-0.03	3.02	0.0006	0.0032	0.0874	0.6769
	6	0.0043	0.0000	0.0000	0.0399	0.7967	0.0214	0.0003	0.0029												
	9	-0.0043	0.0000	0.0000	-0.0399	0.7967	0.0214	0.0003	0.0029												
636	5	1.251	1.03	-0.001	-1.8100	-0.03	3.03	0.05	3.07	0.1263	0.6786	0.0006	-0.0002	0.0001	-1.8100	-0.03	3.03	0.0007	0.0039	0.0902	0.6806
	6	0.0040	0.0000	0.0000	0.0317	0.8509	0.0264	0.0004	0.0035												
	9	-0.0040	0.0000	0.0000	-0.0317	0.8509	0.0264	0.0004	0.0035												
636	6	1.251	3.15	-0.001	-2.8040	-0.02	3.06	0.05	3.10	0.1115	0.6717	0.0003	-0.0001	0.0001	-2.8040	-0.02	3.06	0.0010	0.0066	0.0925	0.6652
	6	0.0034	0.0000	0.0000	0.0067	0.8651	0.0467	0.0008	0.0062												
	9	-0.0034	0.0000	0.0000	-0.0067	0.8651	0.0467	0.0008	0.0062												
636	7	1.249	6.31	-0.001	1.0050	-0.03	3.11	0.05	3.14	0.1077	0.6537	0.0011	-0.0004	0.0000	1.0050	-0.03	3.11	0.0015	0.0096	0.0931	0.6445
	6	0.0011	0.0002	0.0000	0.1794	0.6757	0.0745	0.0013	0.0096												
	9	-0.0011	0.0002	0.0000	-0.1794	0.6757	0.0745	0.0013	0.0096												
636	8	1.252	9.43	-0.001	0.5013	-0.03	3.13	0.04	3.16	0.0984	0.6326	0.0014	-0.0001	0.0000	0.5013	-0.03	3.13	0.0018	0.0121	0.0882	0.6273
	6	0.0044	0.0000	0.0000	0.0843	0.6761	0.0961	0.0016	0.0120												
	9	-0.0044	0.0000	0.0000	-0.0843	0.6761	0.0961	0.0016	0.0120												
636	9	1.249	12.66	-0.001	0.5208	-0.03	3.16	0.04	3.19	0.0937	0.6132	0.0016	-0.0001	0.0000	0.5208	-0.03	3.16	0.0021	0.0138	0.0846	0.6072
	6	0.0016	0.0000	0.0000	0.0789	0.8089	0.1143	0.0019	0.0138												
	9	-0.0016	0.0000	0.0000	-0.0789	0.8089	0.1143	0.0019	0.0138												
636	10	1.250	-0.02	-0.001	-2.5553	-0.03	3.01	0.05	3.07	0.1516	0.6663	0.0005	-0.0002	0.0000	-2.5553	-0.03	3.01	0.0005	0.0024	0.0847	0.6803
	6	0.0005	0.0002	0.0000	0.0642	0.8462	0.0161	0.0002	0.0021												
	9	-0.0005	0.0002	0.0000	-0.0642	0.8462	0.0161	0.0002	0.0021												
637	1	1.251	-3.08	-0.001	1.6108	-0.03	6.01	0.04	6.00	0.2771	0.7459	0.0008	-0.0002	0.0000	1.6108	-0.03	6.01	0.0003	0.0010	0.0501	0.7844
	6	0.0008	0.0002	0.0000	0.0064	0.8721	0.0067	0.0000	0.0009												
	9	-0.0008	0.0002	0.0000	-0.0064	0.8721	0.0067	0.0000	0.0009												
637	2	1.252	-1.03	-0.001	17.2900	-0.03	6.04	0.05	6.03	0.1284	0.6974	0.0002	-0.0002	0.0000	17.2900	-0.03	6.04	0.0006	0.0037	0.0874	0.6943
	6	0.0000	0.0000	0.0000	0.0126	-0.4496	0.0258	0.0004	0.0035												
	9	-0.0000	0.0000	0.0000	-0.0126	-0.4496	0.0258	0.0004	0.0035												
637	3	1.251	0.00	-0.001	-3.1776	-0.03	6.06	0.05	6.04	0.1136	0.6863	0.0004	-0.0002	0.0000	-3.1776	-0.03	6.06	0.0008	0.0051	0.0928	0.6893
	6	0.0004	0.0000	0.0000	0.0240	0.9551	0.0351	0.0006	0.0048												
	9	-0.0004	0.0000	0.0000	-0.0240	0.9551	0.0351	0.0006	0.0048												

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																			
637	4	7	1.251	0.56	-0.00	-3.6712	-0.03	6.07	0.05	6.05	0.1131	0.6856	0.0003	-0.0001	0.0001	-0.03	2.8193	0.429	0.0009	0.0058	0.0918	0.6811
		9	-0.0042	0.0000	-0.0007	-0.0333	0.8992	0.0403	0.0007	0.0054												
637	5	7	1.249	1.08	-0.00	-1.9283	-0.03	6.08	0.05	6.06	0.1092	0.6737	0.0006	-0.0007	0.0001	1.2604	0.484	0.0010	0.0065	0.0902	0.6790	0.6737
		9	-0.0045	0.0000	-0.0007	-0.0567	0.7814	0.0455	0.0008	0.0061												
637	6	7	1.250	3.18	-0.00	-5.7214	-0.02	6.11	0.05	6.09	0.1071	0.6640	0.0001	-0.0001	0.0001	3.6474	0.662	0.0014	0.0087	0.0920	0.6595	0.6595
		9	-0.0037	0.0000	-0.0006	-0.0601	0.8572	0.0642	0.0011	0.0084												
637	7	7	1.252	6.32	-0.00	0.9235	-0.03	6.14	0.05	6.12	0.1001	0.6417	0.0011	-0.0005	0.0000	0.2425	0.882	0.0017	0.0113	0.0875	0.6339	0.6339
		9	-0.0029	0.0000	-0.0005	-0.1488	0.8526	0.0890	0.0015	0.0112												
77	8	7	1.249	9.44	-0.00	0.6833	-0.03	6.17	0.04	6.14	0.0945	0.6216	0.0011	-0.0000	0.0000	0.1793	1.084	0.0020	0.0134	0.0833	0.6216	0.6216
		9	-0.0043	0.0000	-0.0007	0.0973	0.8398	0.1088	0.0018	0.0133												
637	9	7	1.253	12.67	-0.00	0.5948	-0.03	6.19	0.05	6.16	0.0937	0.5903	0.0015	-0.0006	0.0000	0.5684	1.203	0.0022	0.0144	0.0942	0.5903	0.5903
		9	-0.0050	0.0000	-0.0006	0.0430	0.6502	0.1205	0.0022	0.0142												
637	10	7	1.250	0.01	-0.00	-4.3525	-0.03	6.06	0.05	6.04	0.1118	0.6832	0.0001	-0.0007	0.0000	0.903	0.377	0.0008	0.0051	0.0900	0.6809	0.6809
		9	-0.0041	0.0000	-0.0007	-0.0353	0.9521	0.0352	0.0006	0.0048												
63A	1	7	1.251	-3.07	-0.00	2.7345	-0.03	9.09	0.05	9.01	0.1462	0.7124	0.0005	-0.0008	0.0000	0.1968	0.239	0.0007	0.0034	0.0952	0.7124	0.7124
		9	-0.0047	0.0000	-0.0008	-0.0124	0.9335	0.0237	0.0004	0.0033												
63A	2	7	1.252	-0.98	-0.00	19.4473	-0.03	9.12	0.05	9.04	0.1081	0.6901	0.0000	-0.0008	0.0000	0.0047	0.460	0.0009	0.0063	0.0872	0.6901	0.6901
		9	-0.0043	0.0000	-0.0008	-0.0247	0.9515	0.0446	0.0007	0.0061												
63A	3	7	1.250	0.03	-0.00	-4.2872	-0.03	9.14	0.05	9.05	0.1057	0.6608	0.0003	-0.0001	0.0000	0.289	0.556	0.0011	0.0075	0.0892	0.6608	0.6608
		9	-0.0041	0.0000	-0.0008	-0.0183	0.9772	0.0535	0.0009	0.0072												
63A	4	7	1.251	0.57	-0.00	-4.5239	-0.03	9.15	0.05	9.06	0.1059	0.6744	0.0002	-0.0008	0.0000	0.3923	0.604	0.0012	0.0081	0.0901	0.6744	0.6744
		9	-0.0039	0.0000	-0.0008	-0.0082	1.0409	0.0579	0.0010	0.0077												

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key														
638	5	7	1.252	1.10	-0.0002	-0.0001	-0.0007	-0.0000	-2.2512	1.1981	-0.03	9.15	0.05	9.07	0.1047	0.6682	
		8	0.0005	0.0000	0.0000	0.0000	0.0000	0.0000	-0.0277	1.0227	0.653	0.0013	0.0087	0.0904	0.6650		
		9	-0.0038	0.0000	0.0000	-0.0007	-0.0000	-0.0000	0.0626	0.0626	0.0626	0.0011	0.0083	0.0083	0.0083		
638	6	7	1.251	3.22	-0.0002	-0.0000	-0.0006	-0.0000	-2.4205	0.9116	-0.03	9.18	0.05	9.08	0.1002	0.6505	
		8	0.0004	0.0000	0.0000	0.0000	0.0000	0.0000	0.0642	0.9199	0.0815	0.0016	0.0106	0.0914	0.6498		
		9	-0.0036	0.0000	0.0000	-0.0006	-0.0000	-0.0000	0.0789	0.0789	0.0789	0.0014	0.0102	0.0102	0.0102		
638	7	7	1.250	6.35	-0.0002	-0.0000	-0.0004	-0.0000	1.1001	-0.03	9.22	0.05	9.11	0.0941	0.6299		
		8	0.0009	0.0001	0.0000	0.0000	0.0000	0.0000	0.1649	0.2459	0.1017	0.0019	0.0128	0.0845	0.6233		
		9	-0.0033	0.0001	0.0000	-0.0004	-0.0000	-0.0000	0.0960	0.0960	0.0960	0.0017	0.0127	0.0127	0.0127		
638	8	7	1.252	9.45	-0.0001	-0.0000	-0.0007	-0.0000	0.7121	-0.03	9.24	0.04	9.14	0.1041	0.6034		
		8	0.0010	0.0001	0.0000	0.0000	0.0000	0.0000	0.886	0.2518	0.1151	0.0024	0.0139	0.0945	0.5973		
		9	-0.0042	0.0000	0.0000	-0.0007	-0.0000	0.0000	0.0886	0.8386	0.1164	0.0022	0.0139	0.0139	0.0139		
638	9	7	1.252	12.68	-0.0001	-0.0000	-0.0007	-0.0000	0.4926	-0.03	9.25	0.05	9.15	0.0890	0.5926		
		8	0.0015	0.0001	0.0000	0.0001	0.0000	0.0000	0.764	0.472	0.1272	0.0022	0.0150	0.0871	0.5867		
		9	-0.0041	0.0000	0.0000	-0.0007	-0.0000	0.0000	0.0764	0.8472	0.1292	0.0022	0.0151	0.0151	0.0151		
638	10	7	1.250	0.03	-0.0002	-0.0000	-0.0008	-0.0000	8.4291	-0.03	9.14	0.05	9.05	0.1040	0.6762		
		8	0.0001	0.0000	0.0000	0.0000	0.0000	0.0000	0.474	0.7622	0.0557	0.0011	0.0075	0.0890	0.6748		
		9	-0.0042	0.0000	0.0000	-0.0008	-0.0000	0.0000	0.0474	0.9517	0.0534	0.0009	0.0072	0.0890	0.6748		
639	1	7	1.251	3.02	-0.0002	-0.0000	-0.0008	-0.0000	1.8527	-0.03	12.05	0.05	12.05	0.1123	0.6970		
		8	0.0007	0.0000	0.0000	0.0000	0.0000	0.0000	0.476	0.1072	0.0430	0.0009	0.0059	0.0880	0.6948		
		9	-0.0051	0.0000	0.0000	-0.0008	-0.0000	0.0000	0.0476	0.8381	0.0429	0.0007	0.0059	0.0880	0.6948		
639	2	7	1.250	0.95	-0.0002	-0.0000	-0.0008	-0.0000	4.8660	-0.03	12.09	0.05	12.08	0.1040	0.6777		
		8	0.0002	0.0000	0.0000	0.0000	0.0000	0.0000	0.081	0.830	0.0628	0.0013	0.0085	0.0887	0.6723		
		9	-0.0039	0.0000	0.0000	-0.0008	-0.0000	0.0000	0.081	1.0297	0.0621	0.0011	0.0083	0.0887	0.6723		
639	3	7	1.251	0.05	-0.0003	-0.0000	-0.0007	-0.0000	-2.4369	-0.03	12.10	0.05	12.09	0.1024	0.6654		
		8	0.0006	0.0000	0.0000	0.0000	0.0000	0.0000	0.071	0.9063	0.0716	0.0014	0.0093	0.0903	0.6655		
		9	-0.0041	0.0000	0.0000	-0.0007	-0.0000	0.0000	0.071	0.9369	0.0698	0.0012	0.0093	0.0903	0.6655		
639	4	7	1.250	0.59	-0.0002	-0.0000	-0.0007	-0.0000	-5.4148	-0.03	12.11	0.05	12.10	0.1021	0.6624		
		8	0.0002	0.0000	0.0000	0.0000	0.0000	0.0000	0.099	2.7012	0.0758	0.0015	0.0100	0.0903	0.6624		
		9	-0.0040	0.0000	0.0000	-0.0007	-0.0000	0.0000	0.099	0.9486	0.0740	0.0013	0.0097	0.0903	0.6624		
639	5	7	1.251	1.12	-0.0002	-0.0000	-0.0007	-0.0000	-2.5835	-0.03	12.12	0.05	12.10	0.1005	0.6564		
		8	0.0004	0.0000	0.0000	0.0000	0.0000	0.0000	0.229	1.3928	0.0802	0.0016	0.0105	0.0888	0.6564		
		9	-0.0040	0.0000	0.0000	-0.0007	-0.0000	0.0000	0.229	0.9546	0.0781	0.0013	0.0102	0.0888	0.6564		

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																			
639	6	7	1.251	3.25	-0.000	-2.4169	-0.0006	-0.0000	-0.0001	0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	12.14	0.0018	0.0015	12.121	0.0960	0.6378
		8	0.0003	-0.0001	0.0000	-0.0970	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0949	0.0015	0.0015	0.0118	0.0854	0.6355
		9	-0.0039	0.0000	-0.0006	-0.0970	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0935	0.0015	0.0015	0.0118	0.0854	0.6355
639	7	7	1.250	6.37	-0.000	1.2400	-0.0004	-0.0000	-0.0002	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	12.17	0.0021	0.0019	12.13	0.0937	0.6168
		8	-0.0008	-0.0002	-0.0000	-0.1829	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.1125	0.0021	0.0019	0.0138	0.0855	0.6089
		9	-0.0036	0.0001	-0.0004	-0.1829	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.1130	0.0019	0.0019	0.0137	0.0855	0.6089
639	8	7	1.252	9.46	-0.000	0.6599	-0.0007	-0.0000	-0.0001	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	12.18	0.0022	0.0022	12.16	0.0915	0.5946
		8	-0.0010	-0.0001	-0.0000	0.0979	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.1245	0.0022	0.0022	0.0145	0.0901	0.5880
		9	-0.0039	-0.0000	-0.0007	0.0979	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.1245	0.0022	0.0022	0.0146	0.0901	0.5880
639	9	7	1.252	12.69	-0.000	0.4619	-0.0007	-0.0000	-0.0001	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	12.19	0.0022	0.0022	12.17	0.0851	0.5892
		8	-0.0015	-0.0001	-0.0001	0.0774	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.1343	0.0022	0.0022	0.0158	0.0837	0.5770
		9	-0.0042	-0.0000	-0.0007	0.0774	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.1339	0.0022	0.0022	0.0154	0.0837	0.5770
639	10	7	1.251	0.05	-0.000	-7.7982	-0.0007	-0.0000	-0.0002	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	12.10	0.0014	0.0012	12.09	0.1006	0.6635
		8	0.0001	-0.0002	-0.0001	-0.0570	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0716	0.0012	0.0012	0.0095	0.1006	0.6635
		9	-0.0044	0.0000	-0.0007	-0.0570	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0698	0.0012	0.0012	0.0092	0.0891	0.6631
640	3	7	1.252	-3.01	-0.000	1.8300	-0.0007	-0.0000	-0.0002	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	15.10	0.0013	0.0011	15.07	0.1096	0.6908
		8	-0.0007	-0.0000	-0.0000	-0.0047	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0601	0.0013	0.0011	0.0083	0.0902	0.6819
		9	-0.0051	0.0000	-0.0007	-0.0047	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0610	0.0011	0.0011	0.0083	0.0902	0.6819
640	4	7	1.252	-0.94	-0.000	55.5124	-0.0006	-0.0000	-0.0002	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	15.13	0.0016	0.0013	15.09	0.1045	0.6668
		8	-0.0000	-0.0002	-0.0001	-0.0286	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0781	0.0016	0.0013	0.0104	0.1045	0.6668
		9	-0.0044	0.0000	-0.0006	-0.0286	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0784	0.0013	0.0013	0.0103	0.0877	0.6578
640	5	7	1.251	0.07	-0.000	-4.4242	-0.0006	-0.0000	-0.0002	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	15.15	0.0017	0.0014	15.10	0.1019	0.6545
		8	0.0002	-0.0000	-0.0001	-0.0142	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0860	0.0017	0.0014	0.0112	0.1019	0.6545
		9	-0.0045	0.0000	-0.0006	-0.0142	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0855	0.0014	0.0014	0.0111	0.0871	0.6500
640	6	7	1.252	0.61	-0.000	-3.0667	-0.0006	-0.0000	-0.0002	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	15.15	0.0018	0.0015	15.11	0.1005	0.6483
		8	-0.0003	-0.0000	-0.0001	-0.0102	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0898	0.0018	0.0015	0.0116	0.1005	0.6483
		9	-0.0042	0.0000	-0.0006	-0.0102	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0892	0.0015	0.0015	0.0115	0.0859	0.6456
640	7	7	1.252	1.13	-0.000	-2.4585	-0.0007	-0.0000	-0.0001	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	15.15	0.0018	0.0015	15.11	0.0987	0.6432
		8	0.0003	-0.0001	-0.0001	-0.0589	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0937	0.0018	0.0015	0.0120	0.0987	0.6432
		9	-0.0037	-0.0000	-0.0007	-0.0589	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0930	0.0015	0.0015	0.0119	0.0846	0.6403
640	8	7	1.250	3.25	-0.000	-12.8525	-0.0005	-0.0000	-0.0001	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	15.18	0.0020	0.0017	15.13	0.0941	0.6269
		8	0.0000	-0.0001	-0.0000	-0.0654	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.1072	0.0020	0.0017	0.0134	0.0941	0.6269
		9	-0.0042	0.0000	-0.0005	-0.0654	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.1070	0.0017	0.0017	0.0133	0.0818	0.6247

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																		
640	9	1.252	6.37	-0.000	1.2726	-0.03	15.19	0.05	15.16	0.1003	0.6013	0.0008	0.0002	-0.0003	0.3364	0.1173	0.0023	0.0141	0.0986	0.5916
	8	-0.0033	0.0001	-0.0003	-0.1517	0.5217	0.1191	0.0023	0.0141	0.0986	0.5916									
640	10	1.250	9.50	-0.000	1.0376	-0.03	15.21	0.05	15.17	0.0900	0.5931	0.0008	0.0001	-0.0005	0.3189	0.1298	0.0023	0.0154	0.0857	0.5859
	8	-0.0047	-0.0000	-0.0005	0.0470	0.5629	0.1314	0.0022	0.0154	0.0857	0.5859									
640	11	1.254	12.66	0.000	0.4861	-0.03	15.21	0.04	15.18	0.0801	0.5862	0.0014	0.0001	-0.0004	0.3593	0.1367	0.0021	0.0160	0.0774	0.5720
	8	-0.0047	0.0000	-0.0004	-0.0134	0.5155	0.1368	0.0021	0.0156	0.0774	0.5720									
640	12	1.252	0.06	-0.000	-8.3276	-0.03	15.14	0.05	15.12	0.0992	0.6498	0.0001	0.0002	-0.0006	5.6383	0.0861	0.0017	0.0111	0.0848	0.6464
	8	0.0046	0.0000	-0.0006	-0.0595	0.7172	0.0855	0.0014	0.0110	0.0848	0.6464									
641	1	1.250	0.01	-0.000	0.0760	3.04	-0.05	2.94	-0.03	0.4125	0.6222	0.0178	0.0002	0.0013	0.6776	0.0019	0.0001	0.0000	0.4125	0.6222
	8	0.0100	0.0006	0.0013	0.3111	0.6542	0.0009	-0.0002	0.0000	-1.1902	0.4184								-1.1902	0.4184
641	2	1.253	0.00	-0.000	0.0956	6.07	-0.05	6.10	-0.03	0.3916	0.5880	0.0364	0.0006	0.0049	0.6852	0.0019	0.0001	0.0000	0.3916	0.5880
	8	0.0300	0.0010	0.0040	0.1716	0.6776	0.0008	-0.0002	0.0000	-1.2669	0.4226								-1.2669	0.4226
641	3	1.252	0.01	-0.000	0.0978	9.11	-0.05	9.08	-0.04	0.3463	0.5874	0.0540	0.0013	0.0065	0.6909	0.0023	0.0001	0.0000	0.3463	0.5874
	8	0.0490	0.0013	0.0065	0.1415	0.6698	0.0006	-0.0002	0.0000	-1.6020	0.3905								-1.6020	0.3905
642	1	1.252	-3.13	0.000	0.2308	-3.02	2.96	2.94	-3.03	0.0683	0.6978	0.0174	-0.0008	0.0023	0.6593	-0.0093	-0.0001	-0.0013	0.0683	0.6978
	8	-0.0088	0.0005	-0.0010	0.3172	0.5920	-0.0434	-0.0012	-0.0059	0.1404	0.6855								0.1404	0.6855
642	2	1.252	-0.04	0.000	0.2427	-3.02	3.01	2.94	-2.99	0.1609	0.6760	0.0156	-0.0006	0.0011	0.6442	0.0175	0.0005	0.0020	0.1609	0.6760
	8	-0.0096	0.0006	-0.0011	0.3303	0.6073	-0.0140	-0.0007	-0.0020	0.2521	0.7322								0.2521	0.7322
642	3	1.252	0.51	0.000	0.2616	-3.02	3.02	2.94	-2.98	0.1386	0.6667	0.0148	-0.0007	0.0019	0.6653	0.0230	0.0006	0.0030	0.1386	0.6667
	8	-0.0092	0.0006	-0.0012	0.3637	0.6700	-0.0099	-0.0006	-0.0014	0.3194	0.7289								0.3194	0.7289
642	4	1.251	1.04	0.000	0.2653	-3.02	3.03	2.94	-2.98	0.1262	0.6724	0.0146	-0.0007	0.0011	0.6752	0.0287	0.0005	0.0038	0.1262	0.6724
	8	-0.0096	0.0006	-0.0011	0.3377	0.6201	-0.0048	-0.0005	-0.0007	0.5720	0.6025								0.5720	0.6025

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																
642	5	7	1.251	-0.0006	2.09	-0.0019	0.00	0.2222	0.3406	-3.01	0.6453	3.04	0.0391	2.94	0.0009	2.96	0.0052	0.1153	0.6752
		6	-0.0150	0.0006	-0.0006	0.0012	-0.0019	0.3406	0.6473	0.0019	0.0019	0.0019	0.0002	-0.0002	0.0002	0.0001	-0.0001	-0.7228	0.4909
		9	0.0097	0.0006	0.0006	0.0012	0.0012	0.3406	0.6473	0.0019	0.0019	0.0019	0.0002	-0.0002	0.0002	0.0001	-0.0001	-0.7228	0.4909
642	6	7	1.251	-0.0006	3.09	-0.0020	0.00	0.2091	0.3112	-3.01	0.6536	3.06	0.0487	2.94	0.0010	-2.95	0.0065	0.1097	0.6680
		6	-0.0154	0.0006	-0.0006	0.0013	-0.0020	0.3112	0.6173	0.0102	0.0102	0.0102	0.0000	-0.0000	0.0000	0.0012	-0.0012	-0.0115	0.6264
		9	0.0105	0.0006	0.0006	0.0013	0.0013	0.3112	0.6173	0.0102	0.0102	0.0102	0.0000	-0.0000	0.0000	0.0012	-0.0012	-0.0115	0.6264
642	7	7	1.252	-0.0007	4.19	-0.0021	0.00	0.1940	0.3610	-3.02	0.6592	3.07	0.0572	2.94	0.0012	-2.93	0.0076	0.1087	0.6643
		6	-0.0163	0.0007	-0.0007	0.0015	-0.0021	0.3610	0.7290	0.0200	0.0200	0.0200	0.0002	0.0002	0.0002	0.0025	-0.0025	0.0637	0.6315
		9	0.0103	0.0007	0.0007	0.0015	0.0015	0.3610	0.7290	0.0200	0.0200	0.0200	0.0002	0.0002	0.0002	0.0025	-0.0025	0.0637	0.6315
642	8	7	1.252	-0.0007	-0.0006	0.00	0.00	0.2533	0.3231	-3.02	0.6529	3.01	0.0174	2.94	0.0005	-2.99	0.0023	0.1549	0.6602
		6	-0.0154	0.0006	-0.0006	0.0011	-0.0020	0.3231	0.5833	0.0144	0.0144	0.0144	0.0007	-0.0007	0.0007	0.0021	-0.0021	0.2481	0.7315
		9	0.0098	0.0006	0.0006	0.0011	0.0011	0.3231	0.5833	0.0144	0.0144	0.0144	0.0007	-0.0007	0.0007	0.0021	-0.0021	0.2481	0.7315
643	4	7	1.252	-0.0008	-3.12	-0.0024	0.00	0.2244	0.6372	-3.03	0.6717	-0.011	0.0276	2.93	0.0005	-0.07	0.0036	0.0918	0.6666
		6	-0.0182	0.0008	-0.0008	0.0010	-0.0024	0.6372	0.6372	0.0251	0.0251	0.0251	0.0008	-0.0008	0.0008	0.0033	-0.0033	0.1727	0.6734
		9	0.0085	0.0008	0.0008	0.0010	0.0010	0.6372	0.6372	0.0251	0.0251	0.0251	0.0008	-0.0008	0.0008	0.0033	-0.0033	0.1727	0.6734
643	5	7	1.251	-0.0007	-0.0005	0.00	0.00	0.2469	0.6508	-3.03	0.6806	0.0004	0.0006	2.93	0.0002	-0.03	0.0000	2.0583	0.8053
		6	-0.0157	0.0005	-0.0005	0.0012	-0.0020	0.6508	0.6508	0.0006	0.0006	0.0006	0.0002	-0.0002	0.0002	0.0001	-0.0001	-1.8036	0.8426
		9	0.0097	0.0005	0.0005	0.0012	0.0012	0.6508	0.6508	0.0006	0.0006	0.0006	0.0002	-0.0002	0.0002	0.0001	-0.0001	-1.8036	0.8426
643	6	7	1.251	-0.0007	0.49	-0.0020	0.00	0.2399	0.6559	-3.03	0.6559	0.0042	0.0042	2.93	0.0003	-0.03	0.0006	0.3049	0.6623
		6	-0.0154	0.0006	-0.0006	0.0013	-0.0020	0.6559	0.6559	0.0042	0.0042	0.0042	0.0000	-0.0000	0.0000	0.0006	-0.0006	-0.1145	0.7443
		9	0.0099	0.0006	0.0006	0.0013	0.0013	0.6559	0.6559	0.0042	0.0042	0.0042	0.0000	-0.0000	0.0000	0.0006	-0.0006	-0.1145	0.7443
643	7	7	1.252	-0.0006	1.03	-0.0019	0.00	0.2138	0.6528	-3.02	0.6330	0.0099	0.0004	2.93	0.0004	-0.02	0.0013	0.2128	0.6561
		6	-0.0154	0.0005	-0.0005	0.0013	-0.0019	0.6528	0.6528	0.0086	0.0086	0.0086	0.0000	-0.0000	0.0000	0.0011	-0.0011	-0.0088	0.6745
		9	0.0102	0.0005	0.0005	0.0013	0.0013	0.6528	0.6528	0.0086	0.0086	0.0086	0.0000	-0.0000	0.0000	0.0011	-0.0011	-0.0088	0.6745
643	8	7	1.251	-0.0006	2.05	-0.0019	0.00	0.2207	0.6224	-3.02	0.6540	0.0172	0.0172	2.94	0.0005	-0.00	0.0025	0.1476	0.6462
		6	-0.0150	0.0005	-0.0005	0.0013	-0.0019	0.6224	0.6224	0.0172	0.0172	0.0172	0.0002	-0.0002	0.0002	0.0022	-0.0022	0.0598	0.6678
		9	0.0111	0.0005	0.0005	0.0013	0.0013	0.6224	0.6224	0.0172	0.0172	0.0172	0.0002	-0.0002	0.0002	0.0022	-0.0022	0.0598	0.6678
643	9	7	1.250	-0.0006	3.11	-0.0019	0.00	0.2109	0.6537	-3.01	0.6537	0.0301	0.0301	2.94	0.0007	0.00	0.0039	0.1242	0.6540
		6	-0.0152	0.0006	-0.0006	0.0015	-0.0019	0.6537	0.6537	0.0277	0.0277	0.0277	0.0004	-0.0004	0.0004	0.0036	-0.0036	0.0789	0.6513
		9	0.0111	0.0006	0.0006	0.0015	0.0015	0.6537	0.6537	0.0277	0.0277	0.0277	0.0004	-0.0004	0.0004	0.0036	-0.0036	0.0789	0.6513
643	10	7	1.251	-0.0006	4.16	-0.0020	0.00	0.1865	0.6934	-3.01	0.6442	0.0382	0.0382	2.94	0.0009	0.02	0.0051	0.1206	0.6608
		6	-0.0161	0.0006	-0.0006	0.0016	-0.0020	0.6934	0.6934	0.0382	0.0382	0.0382	0.0006	-0.0006	0.0006	0.0050	-0.0050	0.0910	0.6538
		9	0.0121	0.0006	0.0006	0.0016	0.0016	0.6934	0.6934	0.0382	0.0382	0.0382	0.0006	-0.0006	0.0006	0.0050	-0.0050	0.0910	0.6538

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																			
643	11	7	1.251	-0.007	-0.000	0.2233	-3.02	-0.05	2.93	-0.03	0.3817	7	1.251	-0.007	-0.000	0.2233	-3.02	-0.05	2.93	-0.03	0.3817
		8	-0.0160	0.0006	0.0012	0.3171	0.6378	0.0004	0.0001	0.0000	1.6573	8	-0.0160	0.0006	0.0012	0.3171	0.6378	0.0004	0.0001	0.0000	1.6573
		9	0.0097	-0.0006	-0.0012	0.0000	0.6599	0.0004	-0.0002	0.0000	-2.8590	9	0.0097	-0.0006	-0.0012	0.0000	0.6599	0.0004	-0.0002	0.0000	-2.8590
644	1	7	1.252	-3.13	-0.000	-0.5291	0.48	-0.10	-0.49	-0.07	0.6728	7	1.252	-3.13	-0.000	-0.5291	0.48	-0.10	-0.49	-0.07	0.6728
		8	0.0014	-0.0001	0.0003	0.0342	1.1735	-0.0268	-0.0005	-0.0036	0.1057	8	0.0014	-0.0001	0.0003	0.0342	1.1735	-0.0268	-0.0005	-0.0036	0.1057
		9	-0.0079	-0.0000	-0.0011	0.0000	0.7011	-0.0260	-0.0008	-0.0035	0.1647	9	-0.0079	-0.0000	-0.0011	0.0000	0.7011	-0.0260	-0.0008	-0.0035	0.1647
644	2	7	1.250	-0.06	-0.000	-0.2945	0.48	-0.05	-0.49	-0.03	0.5463	7	1.250	-0.06	-0.000	-0.2945	0.48	-0.05	-0.49	-0.03	0.5463
		8	0.0023	-0.0001	0.0004	0.0387	1.0161	0.0013	0.0001	0.0001	0.4495	8	0.0023	-0.0001	0.0004	0.0387	1.0161	0.0013	0.0001	0.0001	0.4495
		9	-0.0072	-0.0000	-0.0010	0.0000	0.7189	0.0000	-0.0002	0.0000	-23.6633	9	-0.0072	-0.0000	-0.0010	0.0000	0.7189	0.0000	-0.0002	0.0000	-23.6633
644	3	7	1.253	0.51	-0.000	-0.2150	0.48	-0.04	-0.49	-0.02	0.6471	7	1.253	0.51	-0.000	-0.2150	0.48	-0.04	-0.49	-0.02	0.6471
		8	0.0023	-0.0000	0.0005	0.0666	1.1033	0.0061	0.0002	0.0007	0.2153	8	0.0023	-0.0000	0.0005	0.0666	1.1033	0.0061	0.0002	0.0007	0.2153
		9	-0.0067	-0.0000	-0.0011	0.0000	0.8143	0.0038	-0.0000	0.0005	-0.1229	9	-0.0067	-0.0000	-0.0011	0.0000	0.8143	0.0038	-0.0000	0.0005	-0.1229
644	4	7	1.250	1.04	-0.000	-0.2319	0.48	-0.03	-0.49	-0.02	0.6424	7	1.250	1.04	-0.000	-0.2319	0.48	-0.03	-0.49	-0.02	0.6424
		8	0.0029	-0.0001	0.0004	0.0495	0.8176	0.0110	0.0003	0.0014	0.1651	8	0.0029	-0.0001	0.0004	0.0495	0.8176	0.0110	0.0003	0.0014	0.1651
		9	-0.0070	-0.0000	-0.0010	0.0000	0.7612	0.0080	0.0000	0.0010	0.0107	9	-0.0070	-0.0000	-0.0010	0.0000	0.7612	0.0080	0.0000	0.0010	0.0107
644	5	7	1.250	2.05	-0.000	-0.1545	0.49	-0.02	-0.49	-0.00	0.6500	7	1.250	2.05	-0.000	-0.1545	0.49	-0.02	-0.49	-0.00	0.6500
		8	0.0027	-0.0000	0.0005	0.0440	0.9317	0.0202	0.0005	0.0026	0.1327	8	0.0027	-0.0000	0.0005	0.0440	0.9317	0.0202	0.0005	0.0026	0.1327
		9	-0.0072	-0.0000	-0.0010	0.0000	0.6989	0.0166	0.0002	0.0021	0.0635	9	-0.0072	-0.0000	-0.0010	0.0000	0.6989	0.0166	0.0002	0.0021	0.0635
644	6	7	1.252	3.11	-0.000	-0.1417	0.49	-0.00	-0.49	0.00	0.6568	7	1.252	3.11	-0.000	-0.1417	0.49	-0.00	-0.49	0.00	0.6568
		8	0.0024	-0.0000	0.0004	0.0375	0.8445	0.0311	0.0004	0.0040	0.1158	8	0.0024	-0.0000	0.0004	0.0375	0.8445	0.0311	0.0004	0.0040	0.1158
		9	-0.0069	-0.0000	-0.0009	0.0000	0.7026	0.0271	0.0004	0.0035	0.0845	9	-0.0069	-0.0000	-0.0009	0.0000	0.7026	0.0271	0.0004	0.0035	0.0845
644	7	7	1.252	4.16	-0.000	-0.2429	0.48	0.01	-0.49	0.02	0.6607	7	1.252	4.16	-0.000	-0.2429	0.48	0.01	-0.49	0.02	0.6607
		8	0.0020	-0.0000	0.0003	0.0377	0.8330	0.0403	0.0009	0.0053	0.1162	8	0.0020	-0.0000	0.0003	0.0377	0.8330	0.0403	0.0009	0.0053	0.1162
		9	-0.0064	-0.0000	-0.0009	0.0000	0.7103	0.0371	0.0007	0.0049	0.0947	9	-0.0064	-0.0000	-0.0009	0.0000	0.7103	0.0371	0.0007	0.0049	0.0947
644	8	7	1.252	-0.03	-0.000	-0.3210	0.48	-0.05	-0.49	-0.03	0.5466	7	1.252	-0.03	-0.000	-0.3210	0.48	-0.05	-0.49	-0.03	0.5466
		8	0.0024	-0.0001	0.0004	0.0118	0.9108	0.0013	0.0001	0.0001	0.4474	8	0.0024	-0.0001	0.0004	0.0118	0.9108	0.0013	0.0001	0.0001	0.4474
		9	-0.0073	-0.0000	-0.0010	0.0000	0.7163	0.0001	-0.0002	-0.0000	-7.4550	9	-0.0073	-0.0000	-0.0010	0.0000	0.7163	0.0001	-0.0002	-0.0000	-7.4550
645	1	7	1.253	-3.15	-0.000	-0.5804	0.48	-0.056	-0.49	0.42	0.6753	7	1.253	-3.15	-0.000	-0.5804	0.48	-0.056	-0.49	0.42	0.6753
		8	0.0016	-0.0001	0.0003	0.0022	0.9368	-0.0299	-0.0006	-0.0040	0.1040	8	0.0016	-0.0001	0.0003	0.0022	0.9368	-0.0299	-0.0006	-0.0040	0.1040
		9	-0.0083	-0.0000	-0.0010	0.0000	0.6557	-0.0232	-0.0008	-0.0032	0.1733	9	-0.0083	-0.0000	-0.0010	0.0000	0.6557	-0.0232	-0.0008	-0.0032	0.1733
645	2	7	1.253	-0.03	-0.000	-0.3424	0.48	-0.051	-0.48	0.46	1.1780	7	1.253	-0.03	-0.000	-0.3424	0.48	-0.051	-0.48	0.46	1.1780
		8	0.0025	-0.0001	0.0004	0.0355	0.8388	-0.0005	0.0001	-0.0001	-0.4971	8	0.0025	-0.0001	0.0004	0.0355	0.8388	-0.0005	0.0001	-0.0001	-0.4971
		9	-0.0073	-0.0000	-0.0010	0.0000	0.7610	0.0022	-0.0001	0.0002	0.3210	9	-0.0073	-0.0000	-0.0010	0.0000	0.7610	0.0022	-0.0001	0.0002	0.3210

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																				
645	3	7	1.253	0.51	-0.00	-0.2940	0.48	-0.50	-0.48	0.47	0.2458	0.6081	0.0028	-0.0001	0.8283	0.0036	0.0001	0.0004	0.0008	0.0008	0.0000	-0.0128	0.6873
645	4	7	1.250	1.02	-0.00	-0.2081	0.48	-0.49	-0.48	0.47	0.1820	0.6404	0.0028	-0.0001	0.8711	0.0082	0.0003	0.0010	0.0013	0.0010	0.0000	0.0375	0.6705
645	5	7	1.252	2.10	-0.00	-0.1466	0.49	-0.48	-0.49	0.49	0.1365	0.6437	0.0027	-0.0004	0.9002	0.0177	0.0004	0.0004	0.0022	0.0025	0.0003	0.0765	0.6475
645	6	7	1.251	3.12	-0.00	-0.1359	0.49	-0.46	-0.48	0.50	0.1217	0.6558	0.0025	-0.0004	0.8651	0.0277	0.0006	0.0036	0.0038	0.0038	0.0005	0.0886	0.6541
645	7	7	1.251	4.19	-0.00	-0.2299	0.48	-0.44	-0.49	0.52	0.1175	0.6577	0.0019	-0.0003	0.8867	0.0374	0.0008	0.0049	0.0052	0.0052	0.0007	0.0923	0.6503
645	8	7	1.251	-0.02	-0.00	-0.2871	0.48	-0.51	-0.48	0.46	-0.1839	1.3410	0.0022	-0.0001	0.9917	-0.0005	-0.0001	-0.0002	-0.0002	-0.0002	-0.0001	-0.3542	0.5675
646	1	7	1.251	-3.12	-0.00	-0.1222	1.01	-0.10	-1.04	0.07	0.1103	0.6797	0.0040	-0.0261	0.7976	-0.0261	-0.0008	-0.0035	-0.0035	-0.0035	-0.0008	0.1654	0.6863
646	2	7	1.251	-0.02	-0.00	-0.0844	1.01	-0.05	-1.04	0.03	0.3901	0.6886	0.0048	-0.0017	0.8013	0.0017	0.0001	0.0002	0.0000	0.0000	0.0002	-7.5863	0.1706
646	3	7	1.252	0.49	-0.00	-0.0634	1.00	-0.04	-1.04	0.02	0.1955	0.6431	0.0050	-0.0001	0.7761	0.0062	0.0002	0.0002	0.0004	0.0004	0.0001	-0.1597	0.6474
646	4	7	1.251	1.02	-0.00	-0.0388	1.01	-0.03	-1.04	0.02	0.1579	0.6473	0.0053	-0.0111	0.7625	0.0111	0.0003	0.0014	0.0010	0.0010	0.0000	-0.0057	0.6577
646	5	7	1.250	2.05	-0.00	-0.0373	1.00	-0.02	-1.04	0.00	0.1285	0.6552	0.0099	-0.0207	0.7319	0.0207	0.0005	0.0021	0.0021	0.0002	0.0644	0.6537	

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																				
646	6	1.250	3.15	-0.007	-0.0046	1.00	-0.00	-1.04	0.00	0.00	0.1138	0.6549	0.0048	-0.0000	0.0007	0.0000	0.0007	0.0004	0.0040	0.0618	0.6420	
	8	0.0095	-0.0001	-0.0014	0.0816	0.7425	0.0267	0.0004	0.0311	0.0004	0.0034	0.0034	0.0048	0.0000	0.0004	0.0004	0.0004	0.0004	0.0004	0.0004	0.0004	0.0004
646	7	1.251	4.16	-0.005	-0.0796	1.00	0.01	-1.03	0.01	0.02	0.1143	0.6594	0.0042	-0.0001	0.0005	0.0007	0.0007	0.0007	0.0048	0.0960	0.6575	
	8	0.0084	-0.0001	-0.0014	0.1029	0.8316	0.0369	0.0007	0.0403	0.0007	0.0048	0.0048	0.0000	0.0000	0.0007	0.0007	0.0007	0.0007	0.0007	0.0007	0.0007	0.0007
646	9	1.252	-0.003	-0.007	-0.0842	1.01	-0.05	-1.04	0.0017	0.03	0.3193	0.6044	0.0049	-0.0001	0.0007	0.0000	0.0000	0.0000	0.0002	0.4340	-1.9370	
	8	0.0098	-0.0001	-0.0014	0.0786	0.7454	0.0000	0.0002	0.0017	0.0002	0.0002	0.0002	0.0002	0.0000	0.0002	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
647	1	1.251	3.13	-0.005	-0.1022	1.00	-1.00	-1.03	0.00	0.92	0.1083	0.6821	0.0034	-0.0001	0.0005	0.0007	0.0007	0.0007	0.0043	0.1866	0.6776	
	8	0.0111	-0.0001	-0.0014	0.0541	0.6682	0.0199	0.0007	0.0322	0.0007	0.0007	0.0007	0.0007	0.0000	0.0007	0.0000	0.0000	0.0000	0.0027	0.0000	0.0000	
647	2	1.250	-0.002	-0.007	-0.1036	1.01	-0.95	-1.03	0.00	0.96	0.1247	0.8928	0.0048	-0.0001	0.0007	0.0000	0.0000	0.0000	0.0004	0.0365	0.7249	
	8	0.0057	-0.0001	-0.0014	0.0758	0.7377	0.0045	0.0000	0.023	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0006	0.0000	0.0000	
647	3	1.250	0.48	-0.007	-0.0942	1.00	-0.94	-1.03	0.00	0.97	0.4169	0.5192	0.0053	-0.0001	0.0007	0.0000	0.0000	0.0001	0.0011	0.0300	0.6722	
	8	0.0098	-0.0001	-0.0014	0.0749	0.7197	0.0086	0.0000	0.014	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0001	0.0011	0.0000	0.0000	
647	4	1.250	1.04	-0.007	-0.0185	1.01	-0.94	-1.03	0.00	0.98	0.1834	0.6278	0.0050	-0.0001	0.0007	0.0000	0.0000	0.0000	0.0007	0.0497	0.6454	
	8	0.0092	-0.0001	-0.0014	0.1024	0.7922	0.0137	0.0001	0.062	0.0001	0.0001	0.0001	0.0001	0.0000	0.0001	0.0000	0.0000	0.0001	0.0017	0.0000	0.0000	
647	5	1.251	2.08	-0.007	-0.0085	1.00	-0.92	-1.03	0.00	0.99	0.1354	0.6375	0.0051	-0.0002	0.0007	0.0000	0.0000	0.0000	0.0019	0.0782	0.6540	
	8	0.0091	-0.0002	-0.0014	0.1095	0.7914	0.0225	0.0003	0.153	0.0003	0.0003	0.0003	0.0003	0.0000	0.0003	0.0000	0.0000	0.0029	0.0029	0.0000	0.0000	
647	6	1.252	3.14	-0.006	-0.0536	1.00	-0.90	-1.03	0.00	1.00	0.1167	0.6436	0.0050	-0.0000	0.0006	0.0000	0.0000	0.0000	0.0032	0.0875	0.6537	
	8	0.0094	-0.0001	-0.0013	0.0887	0.6987	0.0331	0.0005	0.254	0.0005	0.0005	0.0005	0.0005	0.0000	0.0005	0.0000	0.0000	0.0043	0.0043	0.0000	0.0000	
647	7	1.251	4.17	-0.005	-0.0484	1.00	-0.89	-1.03	0.00	1.02	0.1148	0.6516	0.0041	-0.0000	0.0005	0.0000	0.0000	0.0000	0.0045	0.0953	0.6598	
	8	0.0093	-0.0001	-0.0012	0.0742	0.6579	0.0432	0.0008	0.349	0.0008	0.0008	0.0008	0.0008	0.0000	0.0008	0.0000	0.0000	0.0057	0.0057	0.0000	0.0000	
647	8	1.252	-0.005	-0.007	-0.0732	1.01	-0.95	-1.03	0.00	0.96	0.1507	0.9421	0.0047	-0.0000	0.0007	0.0000	0.0000	0.0000	0.0004	0.0531	0.6618	
	8	0.0099	-0.0001	-0.0014	0.0591	0.7192	0.0042	0.0000	0.024	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0005	0.0005	0.0000	0.0000	

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

← See Key →

Run	Pt.	Cd																		
648	3	7	1.249	-3.14	-0.00	2.07	-0.11	-2.12	-0.06	0.1041	0.6700									
		8	0.0096	0.0001	0.0014	0.7457	-0.0260	0.0005	-0.0034	0.1597	0.6708									
		9	-0.0161	-0.0003	-0.0023	0.7160	-0.0262	-0.0008	-0.0035											
648	4	7	1.252	-0.04	-0.00	2.07	-0.06	-2.12	-0.03	0.3916	0.6537									
		8	0.0111	0.0000	0.0015	0.6943	0.0017	0.0001	0.0002	26.8453	-1.8458									
		9	-0.0160	-0.0003	-0.0021	0.6798	-0.0000	-0.0002	0.0000											
648	5	7	1.252	0.00	-0.00	2.07	-0.05	-2.11	-0.02	0.2130	0.6615									
		8	0.0105	0.0001	0.0015	0.7470	0.0060	0.0002	0.0006	-0.1038	0.7312									
		9	-0.0162	-0.0003	-0.0021	0.6660	-0.0034	-0.0000	0.0005											
648	6	7	1.251	0.99	-0.00	2.07	-0.05	-2.11	-0.01	0.1613	0.6489									
		8	0.0111	0.0001	0.0015	0.6960	0.0108	0.0003	0.0014	0.0221	0.7107									
		9	-0.0156	-0.0004	-0.0022	0.7281	0.0075	0.0000	0.0010											
648	7	7	1.252	2.08	-0.00	2.07	-0.02	-2.12	-0.00	0.1280	0.6489									
		8	0.0113	0.0001	0.0015	0.6732	0.0206	0.0005	0.0026	0.0718	0.6647									
		9	-0.0159	-0.0003	-0.0021	0.6888	0.0162	0.0002	0.0021											
648	8	7	1.251	3.13	-0.00	2.07	-0.02	-2.11	0.01	0.1182	0.6594									
		8	0.0105	0.0001	0.0014	0.6808	0.0310	0.0007	0.0041	0.0880	0.6567									
		9	-0.0160	-0.0003	-0.0021	0.6767	0.0264	0.0004	0.0034											
648	9	7	1.251	4.19	-0.00	2.07	-0.00	-2.11	0.02	0.1174	0.6595									
		8	0.0096	0.0001	0.0013	0.6787	0.0405	0.0009	0.0053	0.0981	0.6602									
		9	-0.0153	-0.0003	-0.0021	0.6903	0.0367	0.0007	0.0048											
648	10	7	1.250	-0.07	-0.00	2.07	-0.06	-2.11	-0.03	0.4119	0.5590									
		8	0.0110	0.0000	0.0015	0.6988	0.0014	0.0001	0.0001	2.9794	1.1062									
		9	-0.0161	-0.0003	-0.0021	0.6818	-0.0003	-0.0002	-0.0000											
649	1	7	1.252	-3.13	-0.00	2.07	-1.90	-2.11	1.92	0.1046	0.6809									
		8	0.0093	0.0001	0.0013	0.7223	-0.0375	-0.0007	-0.0018	0.2078	0.6508									
		9	-0.0164	-0.0003	-0.0022	0.6975	-0.0145	-0.0006	-0.0018											
649	2	7	1.251	-0.04	-0.00	2.07	-1.85	-2.11	1.97	0.1410	0.7523									
		8	0.0110	0.0000	0.0014	0.6539	-0.0069	-0.0001	-0.0010	0.0731	0.7016									
		9	-0.0158	-0.0003	-0.0021	0.6791	-0.0091	-0.0001	-0.0012											
649	3	7	1.251	0.51	-0.00	2.07	-1.84	-2.11	1.97	0.1019	0.8943									
		8	0.0105	0.0001	0.0015	0.7251	-0.0025	-0.0000	-0.0004	0.0778	0.6740									
		9	-0.0160	-0.0003	-0.0021	0.6577	-0.0141	-0.0002	-0.0015											

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd																									
649	4	7	8	9	1.251	0.0001	1.01	-0.0003	0.0015	0.0001	0.0003	0.0855	0.1167	2.07	0.7344	0.6773	-1.83	0.0014	0.0184	-2.11	0.0001	0.0025	0.3842	0.0905	0.5130	0.6831
649	5	7	8	9	1.250	0.0001	2.09	-0.0004	0.0015	0.0001	0.0005	0.0634	0.1364	2.07	0.6775	0.7156	-1.82	0.0103	0.0285	-2.11	0.0003	0.0038	0.1573	0.0928	0.6486	0.6747
649	6	7	8	9	1.250	0.0001	3.13	-0.0004	0.0014	0.0001	0.0002	0.0783	0.1360	2.07	0.6907	0.6944	-1.80	0.0200	0.0390	-2.11	0.0005	0.0051	0.1291	0.0943	0.6379	0.6655
649	7	7	8	9	1.251	0.0002	4.18	-0.0003	0.0013	0.0002	0.0003	0.1105	0.1228	2.07	0.7241	0.6507	-1.78	0.0293	0.0488	-2.11	0.0007	0.0065	0.1219	0.0983	0.6432	0.6657
649	8	7	8	9	1.251	0.0000	-0.04	-0.0003	0.0014	0.0000	0.0001	0.0422	0.1020	2.07	0.6917	0.6840	-1.85	0.0069	0.0091	-2.11	0.0001	0.0012	0.1419	0.0662	0.7622	0.6862
650	1	7	8	9	1.252	0.0006	-3.13	-0.0006	0.0039	0.0006	0.0045	0.1125	0.0994	5.06	0.6999	0.6693	-0.11	0.0255	-0.0275	-5.06	0.0008	0.0037	0.1197	0.1492	0.6763	0.6748
650	2	7	8	9	1.252	0.0005	-0.04	-0.0007	0.0039	0.0005	0.0046	0.0947	0.1094	5.06	0.6956	0.6831	-0.06	0.0026	-0.0007	-5.06	0.0001	0.0000	0.1554	0.1368	0.6236	0.5638
650	3	7	8	9	1.252	0.0005	0.50	-0.0008	0.0039	0.0005	0.0047	0.0997	0.1194	5.06	0.6983	0.6995	-0.05	0.0070	0.0027	-5.07	0.0002	0.0003	0.1532	-0.1154	0.6613	0.7093
650	4	7	8	9	1.253	0.0005	1.02	-0.0007	0.0039	0.0005	0.0047	0.0968	0.1144	5.06	0.6916	0.6886	-0.05	0.0121	0.0072	-5.06	0.0003	0.0010	0.1306	0.0388	0.6472	0.6920
650	5	7	8	9	1.252	0.0005	2.08	-0.0008	0.0039	0.0005	0.0047	0.0961	0.1166	5.06	0.6854	0.6916	-0.03	0.0220	0.0161	-5.06	0.0004	0.0021	0.1113	0.0832	0.6473	0.6606
650	6	7	8	9	1.251	0.0005	3.14	-0.0007	0.0037	0.0005	0.0047	0.0951	0.1106	5.05	0.6761	0.6767	-0.02	0.0326	0.0260	-5.07	0.0006	0.0034	0.1038	0.0923	0.6525	0.6537

See Key

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key →									
650	7	1.251	4.19	-0.00	0.1082	5.05	-0.00	-5.06	0.02	0.1107	0.6603
	8	0.0261	0.0005	0.0036	0.1184	0.6897	0.0420	0.0009	0.0055	0.1035	0.6609
	9	-0.0343	-0.0008	-0.0047		0.6901	0.0363	0.0007	0.0048		
650	7	1.250	-0.04	-0.00	0.0954	5.06	-0.06	-5.06	-0.03	0.1495	0.6149
	8	0.0287	0.0005	0.0040	0.1040	0.6968	0.0025	0.0000	0.0003	1.6225	0.9495
	9	-0.0343	-0.0007	-0.0046		0.6730	-0.0006	-0.0002	-0.0001		
651	7	1.250	-3.14	-0.00	0.1342	5.06	-5.12	-5.06	5.00	0.1017	0.6738
	8	0.0266	0.0007	0.0037	0.1106	0.6964	-0.0559	-0.0011	-0.0075	-0.6920	0.9736
	9	-0.0335	-0.0007	-0.0045		0.6837	0.0010	-0.0001	0.0001		
651	7	1.250	-0.03	-0.00	0.1077	5.06	-5.06	-5.07	5.05	0.1282	0.6988
	8	0.0276	0.0005	0.0038	0.1205	0.6984	-0.0246	-0.0006	-0.0034	0.1016	0.6811
	9	-0.0332	-0.0008	-0.0045		0.6831	0.0275	0.0005	0.0037		
651	7	1.252	0.49	-0.00	0.1029	5.05	-5.06	-5.06	5.06	0.1465	0.7135
	8	0.0278	0.0005	0.0038	0.1281	0.6919	-0.0192	-0.0005	-0.0027	0.1037	0.6829
	9	-0.0328	-0.0008	-0.0045		0.6992	0.0322	0.0006	0.0044		
651	7	1.252	1.02	-0.00	0.1150	5.06	-5.05	-5.07	5.07	0.1733	0.7325
	8	0.0278	0.0006	0.0039	0.1287	0.7013	-0.0136	-0.0007	-0.0019	0.0994	0.6782
	9	-0.0329	-0.0008	-0.0045		0.6895	0.0378	0.0007	0.0051		
651	7	1.251	2.10	-0.00	0.1054	5.06	-5.04	-5.07	5.08	0.2662	0.8515
	8	0.0282	0.0005	0.0038	0.1358	0.6859	-0.0047	-0.0009	-0.0063	0.0988	0.6749
	9	-0.0326	-0.0008	-0.0045		0.6951	0.0470	0.0009	0.0063		
651	7	1.251	3.12	-0.00	0.1067	5.06	-5.02	-5.07	5.10	0.0473	0.4536
	8	0.0277	0.0005	0.0038	0.1355	0.6881	0.0030	0.0000	0.0075	0.0966	0.6648
	9	-0.0325	-0.0008	-0.0044		0.6764	0.0569	0.0011	0.0075		
651	7	1.250	4.19	-0.00	0.1123	5.05	-5.00	-5.07	5.11	0.1059	0.5886
	8	0.0268	0.0006	0.0036	0.1457	0.6872	0.0115	0.0002	0.0087	0.0976	0.6591
	9	-0.0316	-0.0009	-0.0043		0.6847	0.0660	0.0012	0.0087		
651	7	1.252	-0.05	-0.00	0.1191	5.06	-5.07	-5.07	5.06	0.1329	0.7103
	8	0.0274	0.0006	0.0039	0.1136	0.7131	-0.0248	-0.0006	-0.0035	0.1022	0.6799
	9	-0.0334	-0.0007	-0.0044		0.6724	0.0270	-0.0005	0.0036		
652	7	1.252	-3.21	45.00	0.1395	-3.06	-3.00	-3.06	-3.02	0.1015	0.6858
	8	0.0434	-0.0012	-0.0058	0.1010	0.6722	-0.0387	-0.0007	-0.0053	0.1499	0.6983
	9	-0.0426	-0.0008	-0.0055		0.6562	-0.0339	-0.0010	-0.0047		

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key															
652	2	1.253	-0.0007	-0.03	44.99	0.1922	-3.02	-2.96	-3.02	-3.003	-2.99	0.1219	0.7115	0.2550	0.0021	-0.0020	0.7206
	7	-0.0193	-0.0005	-0.0024	-0.0029	0.1325	0.6404	-0.0143	-0.0007	-0.0007	-0.0020						
	9	-0.0210	-0.0005	-0.0029	-0.0029	0.1325	0.6997	-0.0143	-0.0007	-0.0007	-0.0020						
652	6	1.250	-0.0012	-0.0057	45.00	0.1392	-3.06	-3.00	-3.07	-3.00	-3.00	0.0949	0.6778	0.1465	0.0052	-0.0046	0.6854
	7	-0.0432	-0.0009	-0.0055	-0.0055	0.1082	0.6646	-0.0390	-0.0009	-0.0009	-0.0052						
	9	-0.0422	-0.0009	-0.0055	-0.0055	0.1082	0.6587	-0.0336	-0.0009	-0.0009	-0.0046						
652	7	1.252	-0.0007	-0.011	45.00	0.1961	-3.02	-2.96	-3.03	-2.98	-2.98	0.1094	0.6973	0.2456	0.0022	-0.0020	0.7026
	7	-0.0194	-0.0005	-0.0028	-0.0028	0.1371	0.6400	-0.0145	-0.0007	-0.0007	-0.0020						
	9	-0.0211	-0.0005	-0.0028	-0.0028	0.1371	0.6819	-0.0145	-0.0007	-0.0007	-0.0020						
652	8	1.251	0.0006	0.43	44.99	0.2184	-3.02	-2.95	-3.03	-2.98	-2.98	0.1253	0.7282	0.2921	0.0018	-0.0016	0.7219
	7	-0.0151	-0.0004	-0.0023	-0.0023	0.1260	0.6107	-0.0113	-0.0006	-0.0006	-0.0016						
	9	-0.0177	-0.0004	-0.0023	-0.0023	0.1260	0.6651	-0.0113	-0.0006	-0.0006	-0.0016						
652	9	1.252	0.0006	0.93	44.99	0.2980	-3.01	-2.95	-3.01	-2.97	-2.97	0.1241	0.7481	0.3504	0.0014	-0.0012	0.7465
	7	-0.0109	-0.0003	-0.0019	-0.0019	0.1168	0.6203	-0.0085	-0.0005	-0.0005	-0.0012						
	9	-0.0144	-0.0003	-0.0019	-0.0019	0.1168	0.6608	-0.0085	-0.0005	-0.0005	-0.0012						
652	10	1.252	2.01	44.99	44.99	0.5100	-3.00	-2.94	-3.00	-2.96	-2.96	0.0912	0.8195	0.9206	0.0004	-0.0004	1.0053
	7	-0.0040	-0.0001	-0.0010	-0.0010	0.0966	0.5245	-0.0024	-0.0004	-0.0004	-0.0004						
	9	-0.0072	-0.0001	-0.0010	-0.0010	0.0966	0.7080	-0.0024	-0.0004	-0.0004	-0.0004						
652	11	1.253	3.04	44.99	44.99	-0.6698	-2.98	-2.93	-2.99	-2.95	-2.95	0.4581	0.3143	0.4581	0.0002	0.0002	0.5148
	7	0.0010	-0.0001	-0.0002	-0.0002	-0.5001	1.4187	0.0019	0.0002	0.0002	0.0002						
	9	-0.0012	-0.0001	-0.0002	-0.0002	-0.5001	1.0924	0.0019	0.0002	0.0002	0.0002						
652	12	1.251	4.10	44.99	44.99	0.0079	-2.97	-2.92	-2.97	-2.95	-2.95	0.1983	0.6102	0.1983	0.0002	0.0002	0.6257
	7	0.0077	0.0003	0.0010	0.0010	0.3312	0.6610	0.0074	0.0002	0.0002	0.0002						
	9	0.0057	0.0003	0.0006	0.0006	0.3312	0.5783	0.0074	0.0001	0.0001	0.0001						
652	13	1.252	-0.0007	-0.0004	44.99	0.1927	-3.02	-2.96	-3.03	-2.98	-2.98	0.1199	0.7144	0.2563	0.0022	-0.0021	0.7330
	7	-0.0194	-0.0004	-0.0028	-0.0028	0.1071	0.6374	-0.0143	-0.0007	-0.0007	-0.0021						
	9	-0.0222	-0.0004	-0.0028	-0.0028	0.1071	0.6283	-0.0143	-0.0007	-0.0007	-0.0021						
653	1	1.252	-3.21	45.00	45.00	0.1601	-0.07	-0.09	0.00	0.00	0.06	0.1016	0.6706	0.2175	0.0022	-0.0022	0.6967
	7	-0.0239	-0.0007	-0.0030	-0.0030	0.1053	0.6454	-0.0161	-0.0007	-0.0007	0.0022						
	9	-0.0240	-0.0005	-0.0031	-0.0031	0.1053	0.6605	-0.0161	-0.0007	-0.0007	0.0022						
653	2	1.252	-0.0005	0.05	44.99	0.7018	-0.03	-0.06	0.04	0.04	0.03	-0.8153	1.2548	-0.8153	0.0001	-0.0001	0.4296
	7	-0.0021	-0.0003	-0.0001	-0.0001	0.0194	0.3553	-0.0004	0.0000	0.0000	0.0001						
	9	-0.0047	-0.0000	-0.0005	-0.0005	0.0194	0.6121	-0.0008	-0.0002	-0.0002	0.0001						

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																					
653	3	1.251	0.98	44.99	-0.1077	-0.02	-0.05	0.05	-0.02	0.0006	0.2610	0.6653	0.0035	-0.0000	0.0002	0.0006	0.0001	1.5834	0.9221	0.0046	0.0002	0.0006	0.7131
653	4	1.252	3.06	44.99	0.1004	0.00	-0.02	0.08	-0.00	0.0004	0.1493	0.6524	0.0173	0.0003	0.0020	0.0023	0.0020	0.1888	0.6816	0.0163	0.0002	0.0024	0.6592
653	5	1.250	6.23	44.99	0.1174	0.03	0.00	0.13	0.02	0.0009	0.1213	0.6591	0.0371	0.0008	0.0053	0.0050	0.0053	0.1516	0.6752	0.0376	0.0007	0.0053	0.6548
653	6	1.250	9.40	44.99	0.1210	0.07	0.03	0.17	0.04	0.0013	0.1209	0.6551	0.0553	0.0013	0.0073	0.0073	0.0079	0.1396	0.6446	0.0554	0.0012	0.0072	0.6513
653	7	1.252	12.56	44.99	0.1183	0.09	0.06	0.21	0.07	0.0016	0.1150	0.6447	0.0706	0.0016	0.0092	0.0092	0.0100	0.1319	0.6523	0.0632	0.0015	0.0105	0.6399
653	8	1.250	0.05	44.99	0.7292	-0.04	-0.06	0.04	-0.03	0.0000	2.9473	-7.4040	-0.0021	-0.0003	-0.0006	-0.0001	-0.0006	0.0093	0.4065	0.0000	0.0003	-0.0001	0.2593
654	1	1.250	-3.24	45.00	0.1361	-3.06	-3.00	-3.06	-3.01	-0.0007	0.0975	0.6783	-0.0435	-0.0011	-0.0056	-0.0057	-0.0056	0.1086	0.6653	-0.0388	-0.0010	-0.0046	0.6932
654	2	1.250	-0.09	44.99	0.1980	-3.02	-2.96	-3.02	-2.98	-0.0007	0.1208	0.7089	-0.0190	-0.0007	-0.0024	-0.0024	-0.0028	0.1254	0.6469	-0.0152	-0.0003	-0.0021	0.7253
654	3	1.251	0.93	44.99	0.2936	-3.01	-2.95	-3.01	-2.97	-0.0013	0.1361	0.7696	-0.0109	-0.0006	-0.0018	-0.0018	-0.0018	0.1052	0.6252	-0.0091	-0.0006	-0.0012	0.7658
654	4	1.251	5.05	44.99	-0.5980	-2.98	-2.93	-2.99	-2.96	0.0003	0.3634	0.3911	0.0010	0.0001	0.0002	0.0003	0.0002	-0.4841	1.5320	0.0012	0.0000	0.0001	0.4272
654	5	1.250	6.19	44.99	0.0998	-2.95	-2.89	-2.94	-2.93	0.0004	0.1373	0.6246	0.0200	0.0004	0.0008	0.0025	0.0026	0.2059	0.6361	0.0205	0.0002	0.0025	0.6254

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Pt. Cd	See Key																													
654	6	7	8	9	1.250	0.0367	0.0410	9.36	0.0013	0.0013	44.99	0.0051	0.0054	0.1300	0.1592	-2.91	0.6620	0.6692	-2.86	0.0395	0.0444	-2.90	0.0010	0.0008	-2.90	0.0050	0.0057	0.1335	0.0924	0.6444	0.6480
654	7	7	8	9	1.251	0.0559	0.0623	12.57	0.0014	0.0017	44.99	0.0073	0.0081	0.1307	0.1403	-2.88	0.6587	0.6503	-2.83	0.0564	0.0654	-2.86	0.0014	0.0011	-2.86	0.0073	0.0085	0.1265	0.0910	0.6462	0.6492
654	8	7	8	9	1.251	-0.0194	-0.0219	-0.10	-0.0007	-0.0004	44.99	-0.0025	-0.0028	0.1946	0.1113	-3.03	0.6466	0.6516	-2.96	-0.0152	-0.0142	-3.02	-0.0003	-0.0007	-2.98	-0.0022	-0.0020	0.1283	0.2512	0.7253	0.7375
655	1	7	8	9	1.252	-0.0167	-0.0185	-3.18	-0.0006	-0.0003	45.00	-0.0020	-0.0023	0.1858	0.0891	0.97	0.6204	0.6452	0.99	-0.0118	-0.0101	0.98	0.0002	-0.0005	0.92	-0.0016	-0.0014	0.0896	0.2895	0.7053	0.7043
655	2	7	8	9	1.254	0.0025	-0.0002	-0.0000	0.0000	0.0002	44.99	0.0005	0.0000	-0.1603	-4.3509	1.00	1.0081	1.00918	1.02	0.0052	0.0052	1.02	0.0002	-0.0000	0.95	0.0007	0.0007	0.2522	0.0738	0.6802	0.7180
655	3	7	8	9	1.253	0.0096	0.0066	0.98	0.0001	0.0004	44.99	0.0014	0.0008	0.0558	0.3208	0.7235	0.7235	0.6445	1.02	0.107	0.1112	1.03	0.0003	0.0000	0.96	0.0014	0.0015	0.1790	0.0343	0.6579	0.6761
655	4	7	8	9	1.251	0.0240	0.0241	3.09	0.0005	0.0007	44.99	0.0032	0.0029	0.1118	0.1639	1.04	1.628	0.6133	1.06	0.230	0.0251	1.06	0.0005	0.0003	0.98	0.0030	0.0032	0.1283	0.0781	0.6647	0.6500
655	5	7	8	9	1.250	0.0435	0.0471	6.25	0.0010	0.0014	44.99	0.0058	0.0062	0.1176	0.1487	1.08	0.6767	0.6594	1.10	0.444	0.0472	1.10	0.0010	0.0009	1.01	0.0058	0.0062	0.1139	0.0964	0.6610	0.6606
655	6	7	8	9	1.251	0.0609	0.0658	9.43	0.0014	0.0018	44.99	0.0080	0.0086	0.1165	0.1403	1.11	1.618	0.6561	1.12	0.613	0.0694	1.14	0.0014	0.0012	1.05	0.0080	0.0089	0.1154	0.0935	0.6543	0.6456
655	7	7	8	9	1.250	0.0759	0.0842	12.61	0.0017	0.0021	44.99	0.0098	0.0107	0.1126	0.1303	1.13	1.481	0.6369	1.15	0.758	0.0878	1.16	0.0016	0.0015	1.07	0.0097	0.0111	0.1119	0.0910	0.6432	0.6352
655	8	7	8	9	1.251	0.0225	-0.0002	-0.0001	-0.0001	0.0002	44.99	0.0004	-0.0000	-0.2002	-4.5165	1.01	0.9567	1.0624	1.02	0.0053	0.0052	1.02	0.0002	-0.0000	0.95	0.0006	0.0007	0.2389	0.0800	0.6487	0.6709

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																			
656	1	1.251	-3.14	44.99	0.3115	3.00	2.98	2.90	3.04	-1.0889	1.4823	0.0044	-0.0002	-0.0003	0.0004	0.0001	-0.0000	-0.0001	-0.0000	-4.4959	-0.4922
656	2	1.252	-0.0003	44.99	0.1075	3.03	3.01	2.94	3.07	0.1674	0.6760	0.0140	0.0005	0.0161	0.0005	0.0021	0.0023	0.0021	0.0023	0.0671	0.6729
656	3	1.255	1.01	44.99	0.0861	3.05	3.02	2.95	3.08	0.1387	0.6773	0.0220	0.0007	0.0222	0.0003	0.0030	0.0032	0.0030	0.0032	0.0767	0.6682
656	4	1.251	3.13	44.99	0.1081	3.07	3.03	2.99	3.10	0.1085	0.6721	0.0364	0.0010	0.0355	0.0006	0.0047	0.0051	0.0047	0.0051	0.1085	0.6721
656	5	1.250	6.28	44.99	0.1118	3.10	3.07	3.04	3.13	0.1485	0.6624	0.0552	0.0017	0.0554	0.0011	0.0073	0.0078	0.0073	0.0078	0.1087	0.6516
656	6	1.251	9.44	44.99	0.1094	3.13	3.10	3.06	3.16	0.0765	0.6488	0.0714	0.0020	0.0715	0.0015	0.0022	0.0103	0.0103	0.0103	0.1099	0.6380
656	7	1.251	12.61	44.99	0.1082	3.16	3.12	3.09	3.18	0.0958	0.6351	0.0950	0.0018	0.0982	0.0017	0.0107	0.0123	0.0107	0.0123	0.1070	0.6261
656	8	1.252	-0.0005	44.99	0.0889	3.04	3.00	2.94	3.07	0.0140	0.6576	0.0106	0.0005	0.0169	0.0002	0.0022	0.0022	0.0022	0.0022	0.1542	0.6673
657	13	1.252	-0.0005	44.99	0.1188	3.04	3.03	3.06	3.01	0.0068	0.7057	0.0068	0.0005	0.0152	0.0006	0.0020	0.0020	0.0020	0.0020	0.2015	0.7004
657	14	1.252	0.03	44.99	0.1211	3.08	3.06	3.10	3.04	0.0320	0.6884	0.0320	0.0009	0.0345	0.0006	0.0050	0.0050	0.0050	0.0050	0.1213	0.6884

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key										
657	15	7	1.252	1.09	44.99	0.1158	0.7109	6.07	6.12	6.04	0.1124	0.6847
		8	0.0399	0.0009	0.0056	0.1349	0.6681	0.0439	0.0009	0.0055	0.0873	0.6791
		9	0.0399	0.0010	0.0053							
657	16	7	1.252	3.21	44.99	0.1047	0.6724	6.09	6.15	6.07	0.1063	0.6781
		8	0.0544	0.0011	0.0073	0.1411	0.6580	0.0530	0.0011	0.0071	0.0960	0.6685
		9	0.0562	0.0015	0.0075							
657	17	7	1.250	6.36	44.99	0.1073	0.6660	6.12	6.19	6.10	0.1078	0.6556
		8	0.0713	0.0015	0.0095	0.1288	0.6330	0.0777	0.0015	0.0093	0.0977	0.6488
		9	0.0768	0.0019	0.0097							
657	18	7	1.251	9.51	44.99	0.1073	0.6473	6.14	6.22	6.12	0.1060	0.6385
		8	0.0855	0.0018	0.0110	0.1259	0.6221	0.0852	0.0018	0.0121	0.0935	0.6290
		9	0.0923	0.0023	0.0114							
657	19	7	1.252	12.69	44.99	0.1064	0.6286	6.16	6.24	6.14	0.1027	0.6256
		8	0.0960	0.0020	0.0120	0.1322	0.5990	0.0962	0.0023	0.0126	0.1102	0.5983
		9	0.1031	0.0027	0.0123							
657	20	7	1.250	0.06	44.99	0.1123	0.7066	6.06	6.10	6.04	0.1104	0.6722
		8	0.0329	0.0007	0.0046	0.1526	0.6697	0.0353	0.0006	0.0047	0.0884	0.6768
		9	0.0312	0.0009	0.0041							
658	1	7	1.253	-3.06	44.99	0.1169	0.6671	9.11	9.03	9.02	0.1304	0.7028
		8	0.0302	0.0007	0.0043	0.1329	0.6741	0.0342	0.0008	0.0044	0.0863	0.6910
		9	0.0257	0.0008	0.0034							
658	2	7	1.252	0.11	44.99	0.1130	0.6990	9.13	9.07	9.04	0.1072	0.6788
		8	0.0508	0.0011	0.0071	0.1329	0.6741	0.0529	0.0010	0.0071	0.0917	0.6763
		9	0.0504	0.0013	0.0067							
658	3	7	1.252	1.12	44.99	0.1076	0.6606	9.14	9.09	9.05	0.1042	0.6733
		8	0.0577	0.0012	0.0079	0.1322	0.6606	0.0617	0.0011	0.0082	0.0952	0.6711
		9	0.0581	0.0015	0.0076							
658	4	7	1.252	3.24	44.99	0.1059	0.6721	9.16	9.12	9.07	0.1018	0.6642
		8	0.0704	0.0014	0.0094	0.1324	0.6422	0.0694	0.0014	0.0092	0.0969	0.6559
		9	0.0731	0.0019	0.0093							
658	5	7	1.252	6.38	44.99	0.1027	0.6508	9.18	9.15	9.09	0.1020	0.6447
		8	0.0856	0.0017	0.0111	0.1424	0.6147	0.0855	0.0018	0.0115	0.0978	0.6293
		9	0.0877	0.0024	0.0107							

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																		
658	6	7	1.251	9.53	44.99	0.1044	9.18	9.20	9.17	9.12	0.0995	0.6309	0.0974	0.0020	0.0122	0.6303	0.0977	0.0019	0.0123	0.0995	0.6309
		8	0.0974	0.0020	0.0122	0.1044	0.6303	0.0977	0.0019	0.0123	0.0995	0.6309	0.0974	0.0020	0.0122	0.6303	0.0977	0.0019	0.0123	0.0995	0.6309
		9	0.1001	0.0026	0.0119	0.1322	0.5946	0.1020	0.0023	0.0121	0.1132	0.5943	0.1001	0.0026	0.0119	0.1322	0.1020	0.0023	0.0121	0.1132	0.5943
658	7	7	1.249	12.71	44.99	0.1008	9.19	9.22	9.19	9.13	0.0938	0.6227	1.070	0.0021	0.0131	0.6150	0.1078	0.0020	0.0134	0.0938	0.6227
		8	0.1070	0.0021	0.0131	0.1008	0.6150	0.1078	0.0020	0.0134	0.0938	0.6227	0.1070	0.0021	0.0131	0.6150	0.1078	0.0020	0.0134	0.0938	0.6227
		9	0.1097	0.0029	0.0125	0.1327	0.5733	0.1115	0.0024	0.0129	0.1097	0.5822	0.1097	0.0029	0.0125	0.5733	0.1115	0.0024	0.0129	0.1097	0.5822
658	8	7	1.251	0.08	44.99	0.1113	9.10	9.13	9.07	9.04	0.1045	0.6747	0.0506	0.0011	0.0070	0.6986	0.0529	0.0011	0.0071	0.1045	0.6747
		8	0.0506	0.0011	0.0070	0.1113	0.6986	0.0529	0.0011	0.0071	0.1045	0.6747	0.0506	0.0011	0.0070	0.6986	0.0529	0.0011	0.0071	0.1045	0.6747
		9	0.0503	0.0013	0.0067	0.1331	0.6661	0.0548	0.0010	0.0073	0.0913	0.6728	0.0503	0.0013	0.0067	0.6661	0.0548	0.0010	0.0073	0.0913	0.6728
659	1	7	1.251	-2.96	44.99	0.1127	15.14	15.12	15.12	15.08	0.1119	0.6738	0.0661	0.0014	0.0091	0.6894	0.0673	0.0012	0.0089	0.1119	0.6738
		8	0.0661	0.0014	0.0091	0.1127	0.6894	0.0673	0.0012	0.0089	0.1119	0.6738	0.0661	0.0014	0.0091	0.6894	0.0673	0.0012	0.0089	0.1119	0.6738
		9	0.0612	0.0015	0.0080	0.1305	0.6603	0.0673	0.0012	0.0089	0.0902	0.6662	0.0612	0.0015	0.0080	0.6603	0.0673	0.0012	0.0089	0.0902	0.6662
659	2	7	1.251	0.18	44.99	0.1038	15.17	15.13	15.15	15.11	0.1006	0.6550	0.0839	0.0017	0.0111	0.6225	0.0841	0.0016	0.0110	0.1006	0.6550
		8	0.0839	0.0017	0.0111	0.1038	0.6225	0.0841	0.0016	0.0110	0.1006	0.6550	0.0839	0.0017	0.0111	0.6225	0.0841	0.0016	0.0110	0.1006	0.6550
		9	0.0802	0.0021	0.0100	0.1367	0.6262	0.0864	0.0016	0.0110	0.0934	0.6413	0.0802	0.0021	0.0100	0.6262	0.0864	0.0016	0.0110	0.0934	0.6413
659	3	7	1.252	1.18	44.99	0.1006	15.18	15.14	15.17	15.11	0.0987	0.6343	0.0894	0.0017	0.0116	0.6328	0.0918	0.0017	0.0116	0.0987	0.6343
		8	0.0894	0.0017	0.0116	0.1006	0.6328	0.0918	0.0017	0.0116	0.0987	0.6343	0.0894	0.0017	0.0116	0.6328	0.0918	0.0017	0.0116	0.0987	0.6343
		9	0.0856	0.0024	0.0103	0.1411	0.6058	0.0918	0.0017	0.0116	0.0959	0.6343	0.0856	0.0024	0.0103	0.6058	0.0918	0.0017	0.0116	0.0959	0.6343
659	4	7	1.252	3.27	44.99	0.0995	15.19	15.15	15.19	15.14	0.0956	0.6450	0.0988	0.0019	0.0125	0.6341	0.0971	0.0018	0.0125	0.0956	0.6450
		8	0.0988	0.0019	0.0125	0.0995	0.6341	0.0971	0.0018	0.0125	0.0956	0.6450	0.0988	0.0019	0.0125	0.6341	0.0971	0.0018	0.0125	0.0956	0.6450
		9	0.0950	0.0026	0.0112	0.1388	0.5937	0.0963	0.0023	0.0116	0.1200	0.6039	0.0950	0.0026	0.0112	0.5937	0.0963	0.0023	0.0116	0.1200	0.6039
659	5	7	1.250	6.42	44.99	0.0943	15.21	15.17	15.21	15.15	0.0895	0.6295	1.108	0.0020	0.0138	0.6242	0.1104	0.0019	0.0139	0.0895	0.6295
		8	0.1108	0.0020	0.0138	0.0943	0.6242	0.1104	0.0019	0.0139	0.0895	0.6295	1.108	0.0020	0.0138	0.6242	0.1104	0.0019	0.0139	0.0895	0.6295
		9	0.1055	0.0028	0.012	0.1354	0.5842	0.1072	0.0024	0.0125	0.1127	0.5870	0.1055	0.0028	0.012	0.5842	0.1072	0.0024	0.0125	0.1127	0.5870
659	6	7	1.251	9.55	44.99	0.1012	15.22	15.19	15.21	15.16	0.0918	0.6091	1.172	0.0023	0.0141	0.6029	0.1170	0.0021	0.0142	0.0918	0.6091
		8	0.1172	0.0023	0.0141	0.1012	0.6029	0.1170	0.0021	0.0142	0.0918	0.6091	1.172	0.0023	0.0141	0.6029	0.1170	0.0021	0.0142	0.0918	0.6091
		9	0.1148	0.0027	0.0130	0.1204	0.5698	0.1165	0.0023	0.0133	0.1024	0.5740	0.1148	0.0027	0.0130	0.5698	0.1165	0.0023	0.0133	0.1024	0.5740
659	7	7	1.252	12.71	44.99	0.0943	15.23	15.20	15.22	15.16	0.0909	0.5966	1.233	0.0023	0.0146	0.5939	0.1223	0.0022	0.0145	0.0909	0.5966
		8	0.1233	0.0023	0.0146	0.0943	0.5939	0.1223	0.0022	0.0145	0.0909	0.5966	1.233	0.0023	0.0146	0.5939	0.1223	0.0022	0.0145	0.0909	0.5966
		9	0.1168	0.0026	0.0131	0.1143	0.5641	0.1199	0.0023	0.0134	0.0985	0.5626	0.1168	0.0026	0.0131	0.5641	0.1199	0.0023	0.0134	0.0985	0.5626
659	8	7	1.251	0.14	44.99	0.1037	15.17	15.13	15.15	15.11	0.0977	0.6496	0.0802	0.0017	0.0111	0.6834	0.0842	0.0016	0.0109	0.0977	0.6496
		8	0.0802	0.0017	0.0111	0.1037	0.6834	0.0842	0.0016	0.0109	0.0977	0.6496	0.0802	0.0017	0.0111	0.6834	0.0842	0.0016	0.0109	0.0977	0.6496
		9	0.0802	0.0021	0.0100	0.1368	0.6241	0.0862	0.0016	0.0110	0.0934	0.6403	0.0802	0.0021	0.0100	0.6241	0.0862	0.0016	0.0110	0.0934	0.6403

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																						
660	1	7	1.050	-3.13	0.00	1.0344	-0.03	-3.00	0.05	-3.02	0.1318	0.6734	7	1.051	-0.001	0.0001	1.0344	-0.03	-3.00	0.05	-3.02	0.1318	0.6734	
			-0.0010	-0.0002	-0.0007	0.0330	0.5727	-0.0450	-0.0011	-0.0060	0.1698	0.6740												
			-0.0042	-0.0000	-0.0007	0.0330	0.8631	-0.0437	-0.0014	-0.0058														
660	2	7	1.050	-1.08	0.00	120.7062	-0.03	-2.97	0.05	-2.99	0.1444	0.6745	7	1.050	-0.0002	0.0001	120.7062	-0.03	-2.97	0.05	-2.99	0.1444	0.6745	
			-0.0000	-0.0002	-0.0005	-0.0090	-6.1063	-0.0231	-0.0008	-0.0031	0.2088	0.6750												
			-0.0043	0.0000	-0.0005	-0.0090	0.6701	-0.0231	-0.0009	-0.0031														
660	3	7	1.051	-0.06	0.00	-4.4649	-0.03	-2.96	0.05	-2.99	0.1392	0.7125	7	1.051	-0.001	0.0000	-4.4649	-0.03	-2.96	0.05	-2.99	0.1392	0.7125	
			0.0001	-0.0001	0.0000	0.0381	2.8040	-0.0132	-0.0003	-0.0018	0.2546	0.6920												
			-0.0037	-0.0000	-0.0006	0.0381	0.8167	-0.0143	-0.0007	-0.0019														
660	4	7	1.051	0.48	0.00	-4.6287	-0.03	-2.95	0.05	-2.98	0.1165	0.7217	7	1.051	0.0001	0.0000	-4.6287	-0.03	-2.95	0.05	-2.98	0.1165	0.7217	
			0.0001	-0.0000	-0.0006	-0.0748	4.6636	-0.0089	-0.0002	-0.0012	0.3137	0.7193												
			-0.0034	-0.0000	-0.0006	-0.0748	0.9527	-0.0100	-0.0006	-0.0014														
660	5	7	1.048	1.00	0.00	-1.2731	-0.03	-2.94	0.05	-2.97	0.0695	0.8333	7	1.048	0.0001	0.0000	-1.2731	-0.03	-2.94	0.05	-2.97	0.0695	0.8333	
			0.0004	-0.0001	-0.0006	0.0565	0.8193	-0.0043	-0.0000	-0.0007	0.4056	0.7491												
			-0.0036	-0.0000	-0.0006	0.0565	0.8743	-0.0061	-0.0005	-0.0009														
660	6	7	1.050	3.08	0.00	-3.1393	-0.04	-2.91	0.05	-2.95	0.2470	0.6031	7	1.050	0.0001	0.0000	-3.1393	-0.04	-2.91	0.05	-2.95	0.2470	0.6031	
			0.0001	-0.0001	-0.0006	0.0305	1.6749	0.0112	0.0005	0.0013	0.0796	0.6133												
			-0.0038	-0.0000	-0.0006	0.0305	0.7892	0.0081	0.0001	0.0010														
660	7	7	1.043	6.17	0.00	1.2773	-0.03	-2.87	0.05	-2.91	0.1714	0.6383	7	1.043	0.0000	0.0000	1.2773	-0.03	-2.87	0.05	-2.91	0.1714	0.6383	
			0.0008	-0.0002	-0.0004	-0.0972	0.2161	0.0391	0.0013	0.0050	0.1435	0.6394												
			-0.0034	0.0000	-0.0004	-0.0972	0.6003	0.0382	0.0010	0.0048														
660	8	7	1.052	9.30	0.00	1.5581	-0.03	-2.82	0.04	-2.87	0.1318	0.6384	7	1.052	0.0000	0.0000	1.5581	-0.03	-2.82	0.04	-2.87	0.1318	0.6384	
			0.0008	-0.0002	-0.0005	0.1101	0.4335	0.0708	0.0018	0.0090	0.1162	0.6402												
			-0.0042	-0.0000	-0.0005	0.1101	0.6788	0.0699	0.0016	0.0089														
660	9	7	1.050	12.52	0.00	0.6616	-0.04	-2.78	0.04	-2.85	0.1180	0.6173	7	1.050	0.0001	0.0000	0.6616	-0.04	-2.78	0.04	-2.85	0.1180	0.6173	
			0.0015	-0.0002	-0.0001	0.1981	0.3545	0.0969	0.0022	0.0119	0.1033	0.6238												
			-0.0038	-0.0001	-0.0006	0.1981	0.8831	0.0974	0.0020	0.0121														
660	10	7	1.050	0.06	0.00	-2.6822	-0.03	-2.96	0.05	-2.99	0.1443	0.7277	7	1.050	0.0003	0.0002	-2.6822	-0.03	-2.96	0.05	-2.99	0.1443	0.7277	
			0.0003	-0.0002	-0.0006	0.0208	0.6859	-0.0131	-0.0003	-0.0019	0.2546	0.6973												
			-0.0039	-0.0000	-0.0006	0.0208	0.7667	-0.0143	-0.0007	-0.0019														
661	1	7	1.051	-3.12	0.00	1.4329	-0.03	-1.01	0.05	-1.05	0.1476	0.6686	7	1.051	0.0008	-0.0002	1.4329	-0.03	-1.01	0.05	-1.05	0.1476	0.6686	
			-0.0008	-0.0000	-0.0006	0.0085	0.8929	-0.0308	-0.0009	-0.0041	0.1958	0.6610												
			-0.0046	-0.0000	-0.0006	0.0085	0.7035	-0.0303	-0.0011	-0.0041														

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key														
661	2	7 8 9	1.050 -0.0004 -0.00040	-1.07 -0.0001 -0.0000	-0.00 0.0000 -0.00006	1.6150 0.0246	-0.3663 0.7487	-0.03 -0.0104 -0.0108	-0.97 -0.0104 -0.0108	0.05 -0.0003 -0.0006	-1.03 -0.0015 -0.0014	0.1518 0.2990	0.7220 0.6816				
661	3	7 8 9	1.049 0.0002 -0.00038	-0.06 -0.0001 -0.0000	-0.00 0.0000 -0.00005	-3.1875 0.0391	-0.02 1.3583 0.7571	-0.96 -0.0022 -0.0034	0.05 -0.0000 -0.0004	-1.01 -0.0004 -0.0005	0.0787 0.6346	0.9970 0.7379					
661	4	7 8 9	1.050 -0.0001 -0.00038	0.48 -0.0000 -0.0000	-0.00 0.0001 -0.00005	2.0990 0.0373	-0.03 -3.682 0.7635	-0.95 0.0015 -0.0003	0.05 0.0001 -0.0002	-1.01 0.0000 -0.0000	0.4826 3.1855	0.2077 0.8629					
661	5	7 8 9	1.049 0.0001 -0.00038	1.00 -0.0000 -0.0000	-0.00 0.0001 -0.00005	-2.7674 0.0384	-0.03 3.9076 0.7573	-0.95 0.0056 0.0030	0.05 0.0003 -0.0000	-1.00 0.0006 0.0003	0.3076 -0.1417	0.6058 0.6427					
661	9	7 8 9	1.050 -0.0000 -0.00036	3.09 -0.0000 -0.0000	-0.00 0.0000 -0.00005	5.7765 0.0252	-0.02 -4.5760 0.8172	-0.90 0.0238 0.0197	0.05 0.0009 0.0005	-0.97 0.0029 0.0024	0.1932 0.1321	0.6265 0.6219					
661	10	7 8 9	1.050 -0.0011 -0.00036	6.19 -0.0001 -0.0000	-0.00 -0.0004 -0.00005	0.7932 -0.0937	-0.02 0.1506 0.5608	-0.86 0.0532 0.0519	0.05 0.0013 0.0013	-0.93 0.0069 0.0066	0.1476 0.1280	0.6506 0.6441					
661	11	7 8 9	1.052 -0.0009 -0.00044	9.33 -0.0002 -0.0000	-0.00 -0.0000 -0.00005	1.2388 0.0330	-0.02 0.3939 0.5667	-0.82 0.0820 0.0821	0.04 0.0020 0.0018	-0.90 0.0103 0.0104	0.1270 0.1097	0.6293 0.6335					
661	12	7 8 9	1.047 -0.0014 -0.00044	12.54 -0.0002 -0.0000	-0.00 -0.0000 -0.00005	0.7944 0.0756	-0.03 0.3155 0.6001	-0.79 0.1041 0.1034	0.04 0.0025 0.0023	-0.86 0.0126 0.0124	0.1207 0.1146	0.6063 0.6019					
661	13	7 8 9	1.049 -0.0001 -0.00038	-0.05 -0.0001 -0.0000	-0.00 0.0000 -0.00005	4.9074 -0.0261	-0.02 -3.2326 0.7219	-0.95 -0.0019 -0.0031	0.05 -0.0004 -0.0004	-1.01 0.0004 -0.0005	0.1062 0.6909	1.1163 0.8453					
662	1	7 8 9	1.048 -0.0014 -0.00045	-3.10 -0.0001 -0.0000	0.00 -0.0000 -0.00006	0.5580 -0.0314	-0.03 0.3251 0.6643	-0.11 -0.0247 -0.0232	0.05 -0.0007 -0.0010	-0.06 -0.0032 -0.0031	0.1522 0.2267	0.6616 0.6736					
662	2	7 8 9	1.052 -0.0003 -0.00039	-1.05 -0.0001 -0.0000	-0.00 0.0000 -0.00006	2.6214 0.0034	-0.02 -0.3168 -0.7759	-0.08 -0.0054 -0.0049	0.05 -0.0001 -0.0005	-0.04 -0.0007 -0.0007	0.1292 0.5115	0.7593 0.7563					

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key											
662	3	7 8 9	1.051 0.0000 -0.00040	-0.003 -0.0001 0.0000	-0.000 0.0000 -0.00005	-0.0256 208.0807 0.6725	-0.02 0.0023 0.0012	-0.06 0.0001 -0.0001	0.05 0.0001 -0.0001	0.0002 0.0001 -0.0001	0.3979 0.7689 -0.0001	0.5058 0.5611 0.0002		
662	4	7 8 9	1.050 -0.0000 -0.00037	0.47 -0.0001 -0.0000	0.000 0.0000 -0.00005	15.8798 -12.0835 0.7695	-0.02 0.0023 0.0042	-0.05 0.0063 0.0000	0.05 0.0003 -0.0000	0.0007 0.0005 -0.0005	0.2626 0.0265 -0.0005	0.5996 0.6697 0.0007		
662	5	7 8 9	1.051 0.0002 -0.00040	0.99 -0.0001 0.0000	0.000 0.0000 -0.00005	-2.5023 1.4865 0.6817	-0.02 1.4865 0.6817	-0.04 0.0106 0.0081	0.05 0.0004 0.0001	0.0013 0.0010 0.0010	0.2303 0.0670 0.0000	0.6181 0.6364 0.0013		
662	6	7 8 9	1.050 -0.0000 -0.00036	3.10 -0.0001 -0.0000	0.000 0.0000 -0.00005	7.6937 -4.8983 0.7695	-0.02 0.0301 0.0264	-0.01 0.0010 0.0007	0.05 0.0010 0.0007	0.01 0.0038 0.0033	0.1800 0.1357 0.0000	0.6403 0.6347 0.0038		
662	7	7 8 9	1.053 -0.0008 -0.00036	6.20 -0.0002 0.0000	0.000 -0.0000 -0.00004	1.4763 0.4289 0.5751	-0.03 0.0390 0.0587	0.02 0.0016 0.0014	0.05 0.0016 0.0014	0.04 0.0076 0.0075	0.1400 0.1210 0.0000	0.6493 0.6448 0.0076		
662	8	7 8 9	1.051 -0.0012 -0.00042	9.34 -0.0002 -0.0000	0.000 -0.0000 -0.00005	0.6207 0.6404 0.6404	-0.02 0.0666 0.0875	0.07 0.0021 0.0019	0.04 0.0021 0.0019	0.08 0.0108 0.0109	0.1267 0.1085 0.0000	0.6244 0.6249 0.0108		
662	9	7 8 9	1.051 -0.0017 -0.00036	12.55 -0.0001 -0.0001	0.000 -0.0000 -0.00006	0.5076 0.1695 0.8655	-0.03 0.2311 0.8655	0.09 0.1050 0.1068	0.04 0.0025 0.0023	0.10 0.0125 0.0127	0.1231 0.1110 0.0000	0.5985 0.5946 0.0125		
662	10	7 8 9	1.053 -0.0001 -0.00040	-0.01 -0.0001 0.0000	0.000 -0.0000 -0.00005	5.8362 -3.5024 0.6850	-0.02 0.0026 0.0012	0.06 0.0001 -0.0001	0.05 0.0001 -0.0001	0.02 0.0002 -0.0001	0.3394 -0.7362 0.0000	0.4356 0.4321 0.0002		
663	1	7 8 9	1.052 -0.0012 -0.00045	-3.09 -0.0001 0.0000	0.000 -0.0000 -0.00006	0.7565 0.3494 0.6699	-0.02 0.3494 0.6699	0.97 -0.0172 -0.0165	0.05 0.0005 -0.0008	0.93 -0.0223 -0.0022	0.1602 0.2593 0.0000	0.6707 0.6716 0.0005		
663	2	7 8 9	1.050 -0.0003 -0.00039	-1.04 -0.0001 0.0000	0.000 -0.0000 -0.00005	2.2362 -0.3378 0.7327	-0.02 0.3378 0.7327	1.00 0.0006 0.0000	0.04 0.0000 -0.0002	0.95 -0.0000 -0.0000	0.4816 -0.3919 -4.8070	0.0616 -0.8070 0.0000		
663	3	7 8 9	1.051 -0.0002 -0.00040	-0.03 -0.0001 0.0000	0.000 -0.0001 -0.00005	-0.02 -2.2777 0.6739	-0.02 0.0082 0.0060	1.01 0.0082 0.0060	0.04 0.0003 0.0000	0.97 0.0010 0.0008	0.2360 0.0245 0.0000	0.6194 0.6661 0.0008		

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key															
663	4	7	1.051	0.47	-0.00	-0.001	-3.6989	-0.0104	2.9533	-0.02	1.02	0.05	0.97	0.2195	0.6300		
		8	0.0001	-0.0001	0.0001	0.0001	-0.0104	0.7210	0.0100	0.0123	0.0005	0.0015	0.0687	0.6297			
		9	-0.0040	0.0000	-0.0005	-0.0005	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000			
663	5	7	1.050	1.02	-0.00	0.0001	1.6786	-0.0020	-0.02	1.03	0.05	0.97	0.1995	0.6321			
		8	-0.0001	-0.0000	0.0001	0.0001	0.0000	-5.0625	0.0171	0.0171	0.0006	0.0021	0.1054	0.6422			
		9	-0.0038	-0.0000	-0.0005	-0.0005	0.0000	0.7824	0.0138	0.0138	0.0002	0.0017	0.0000	0.0000			
663	6	7	1.053	3.15	-0.00	0.0000	16.9028	-0.0252	-0.02	1.06	0.05	1.00	0.1666	0.6526			
		8	-0.0000	-0.0001	0.0000	0.0000	0.0000	-5.7424	0.0377	0.0377	0.0012	0.0049	0.1309	0.6415			
		9	-0.0035	-0.0000	-0.0005	-0.0005	0.0000	0.8172	0.0334	0.0334	0.0008	0.0042	0.0000	0.0000			
663	7	7	1.049	6.26	-0.00	0.0000	0.8343	-0.03	-0.03	1.10	0.05	1.04	0.1351	0.6476			
		8	-0.0012	-0.0002	0.0000	0.0000	0.0000	0.2501	0.0665	0.0665	0.0017	0.0086	0.1164	0.6428			
		9	-0.0032	0.0000	-0.0004	-0.0004	-0.0906	0.6822	0.0662	0.0662	0.0015	0.0085	0.0000	0.0000			
663	8	7	1.051	9.35	-0.00	0.0000	0.7865	0.1656	-0.03	1.14	0.04	1.07	0.1278	0.6187			
		8	-0.0012	-0.0001	0.0000	0.0000	0.0000	0.1656	0.0914	0.0914	0.0023	0.0113	0.1147	0.6163			
		9	-0.0039	-0.0000	-0.0005	-0.0005	0.0709	0.6872	0.0912	0.0912	0.0020	0.0112	0.0000	0.0000			
663	9	7	1.053	12.55	-0.00	0.0000	0.6622	0.3209	-0.03	1.16	0.04	1.09	0.1212	0.5936			
		8	-0.0015	-0.0002	0.0000	0.0000	0.0468	0.6122	0.3209	0.1078	0.0026	0.0128	0.1123	0.5903			
		9	-0.0043	-0.0000	-0.0005	-0.0005	0.0000	0.6122	0.1097	0.1097	0.0024	0.0129	0.0000	0.0000			
663	10	7	1.051	-0.04	-0.00	0.0000	-5.4955	-0.5593	-0.02	1.02	0.05	0.96	0.2201	0.6314			
		8	0.0001	-0.0001	0.0000	0.0000	-0.0686	2.9501	0.003	0.003	0.0003	0.0009	0.0266	0.6314			
		9	-0.0044	0.0000	-0.0005	-0.0005	0.0000	0.5593	0.0060	0.0060	0.0000	0.0007	0.0000	0.0000			
664	1	7	1.050	-3.08	0.00	0.0000	0.7623	0.2604	-0.03	2.96	0.05	3.03	0.1686	0.7399			
		8	-0.0011	-0.0001	0.0000	0.0000	-0.0256	0.6417	-0.03	2.96	-0.0001	-0.0008	0.6535	0.7410			
		9	-0.0045	0.0000	-0.0005	-0.0005	0.0000	0.6417	-0.0040	-0.0040	-0.0005	-0.0005	0.0000	0.0000			
664	2	7	1.049	-1.03	-0.00	0.0000	4.0996	-0.8830	-0.02	2.99	0.05	3.05	0.2106	0.6396			
		8	-0.0002	-0.0001	0.0000	0.0000	-0.0172	0.6931	0.0108	0.0108	0.0004	0.0013	0.0688	0.6557			
		9	-0.0040	0.0000	-0.0005	-0.0005	0.0000	0.6931	0.0101	0.0101	0.0001	0.0013	0.0000	0.0000			
664	3	7	1.047	-0.03	-0.00	0.0000	2.8323	-2.7294	-0.02	3.00	0.04	3.06	0.1850	0.6481			
		8	-0.0001	-0.0001	0.0001	0.0001	-0.0087	0.7131	0.0199	0.0199	0.0007	0.0025	0.1078	0.6455			
		9	-0.0039	0.0000	-0.0005	-0.0005	0.0000	0.7131	0.0181	0.0181	0.0003	0.0023	0.0000	0.0000			
664	4	7	1.049	0.52	-0.00	0.0000	5.0536	-7.3010	-0.02	3.01	0.05	3.07	0.1760	0.6522			
		8	-0.0000	-0.0000	0.0001	0.0001	-0.0016	0.7407	0.0252	0.0252	0.0008	0.0029	0.1177	0.6586			
		9	-0.0038	0.0000	-0.0005	-0.0005	0.0000	0.7407	0.0228	0.0228	0.0005	0.0029	0.0000	0.0000			

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																	
664	5	7	8	9	1.052	-0.0002	1.04	-0.000	-0.000	-1.8572	1.7177	-0.02	3.02	0.0302	0.0010	0.05	3.07	0.1686	0.6547
					0.0002	0.0000	-0.0000	0.0005	-0.0314	0.6019	0.0269	0.0006	0.0035	0.1248	0.0035	0.0006	0.0035	0.1248	0.6514
					-0.0042	0.0000	0.0000	-0.0005	-0.0314	0.6019	0.0269	0.0006	0.0035	0.1248	0.0035	0.0006	0.0035	0.1248	0.6514
664	6	7	8	9	1.049	-0.0001	3.19	-0.000	4.7266	-2.7068	0.0512	0.05	3.10	0.1454	0.0067	0.05	3.10	0.1454	0.6634
					0.0001	0.0000	0.0000	-0.0004	-0.0339	0.6012	0.0477	0.0011	0.0062	0.1232	0.0062	0.0011	0.0062	0.1232	0.6536
					-0.0041	0.0000	0.0000	-0.0004	-0.0339	0.6012	0.0477	0.0011	0.0062	0.1232	0.0062	0.0011	0.0062	0.1232	0.6536
664	7	7	8	9	1.050	-0.0013	6.27	-0.000	0.7683	0.2772	3.09	0.05	3.14	0.1292	0.0098	0.05	3.14	0.1292	0.6357
					0.0013	0.0000	0.0000	-0.0003	-0.1300	0.4687	0.0788	0.0017	0.0100	0.1124	0.0100	0.0017	0.0100	0.1124	0.6386
					-0.0039	0.0000	0.0000	-0.0003	-0.1300	0.4687	0.0788	0.0017	0.0100	0.1124	0.0100	0.0017	0.0100	0.1124	0.6386
664	8	7	8	9	1.047	-0.0012	9.37	-0.000	0.8198	-0.2739	3.12	0.04	3.16	0.1308	0.0119	0.04	3.16	0.1308	0.6063
					0.0012	0.0000	0.0000	-0.0005	0.1017	0.7452	0.0980	0.0024	0.0117	0.1225	0.0117	0.0024	0.0117	0.1225	0.5982
					-0.0038	0.0000	0.0000	-0.0005	0.1017	0.7452	0.0980	0.0024	0.0117	0.1225	0.0117	0.0024	0.0117	0.1225	0.5982
664	9	7	8	9	1.052	-0.0015	12.56	-0.000	0.6573	-0.3043	3.14	0.04	3.18	0.1183	0.0132	0.04	3.18	0.1183	0.5877
					0.0015	0.0000	0.0000	-0.0005	0.0790	0.6421	0.1148	0.0025	0.0133	0.1098	0.0133	0.0025	0.0133	0.1098	0.5801
					-0.0042	0.0000	0.0000	-0.0005	0.0790	0.6421	0.1148	0.0025	0.0133	0.1098	0.0133	0.0025	0.0133	0.1098	0.5801
664	10	7	8	9	1.050	-0.0004	0.01	-0.000	0.9769	-0.4787	3.00	0.05	3.06	0.1816	0.0026	0.05	3.06	0.1816	0.6395
					0.0004	0.0000	0.0000	-0.0005	-0.0051	0.7535	0.203	0.0007	0.0023	0.1079	0.0023	0.0007	0.0023	0.1079	0.6360
					-0.0038	0.0000	0.0000	-0.0005	-0.0051	0.7535	0.203	0.0007	0.0023	0.1079	0.0023	0.0007	0.0023	0.1079	0.6360
665	1	7	8	9	1.050	-0.0009	-3.09	-0.000	1.0284	0.2402	6.02	0.04	6.00	0.1970	0.0013	0.04	6.00	0.1970	0.6698
					0.0009	0.0000	0.0000	-0.0006	-0.0229	0.7215	0.103	0.0004	0.0014	0.0485	0.0014	0.0004	0.0014	0.0485	0.6850
					-0.0043	0.0000	0.0000	-0.0006	-0.0229	0.7215	0.103	0.0004	0.0014	0.0485	0.0014	0.0004	0.0014	0.0485	0.6850
665	2	7	8	9	1.053	-0.0004	-1.02	-0.000	1.9729	-0.6994	6.05	0.05	6.03	0.1636	0.0039	0.05	6.03	0.1636	0.6738
					0.0004	0.0000	0.0000	-0.0005	-0.0346	0.6617	0.296	0.0009	0.0037	0.1118	0.0037	0.0009	0.0037	0.1118	0.6553
					-0.0042	0.0000	0.0000	-0.0005	-0.0346	0.6617	0.296	0.0009	0.0037	0.1118	0.0037	0.0009	0.0037	0.1118	0.6553
665	3	7	8	9	1.051	-0.0002	0.00	-0.000	-3.2510	1.0530	6.06	0.05	6.05	0.1511	0.0054	0.05	6.05	0.1511	0.6732
					0.0002	0.0000	0.0000	-0.0005	-0.0417	0.5978	0.402	0.0012	0.0050	0.1162	0.0050	0.0012	0.0050	0.1162	0.6626
					-0.0043	0.0000	0.0000	-0.0005	-0.0417	0.5978	0.402	0.0012	0.0050	0.1162	0.0050	0.0012	0.0050	0.1162	0.6626
665	4	7	8	9	1.049	-0.0000	0.52	-0.000	6.2136	-0.4275	6.07	0.05	6.04	0.1430	0.0061	0.05	6.04	0.1430	0.6694
					0.0000	0.0000	0.0000	-0.0005	0.0071	0.7632	0.460	0.0013	0.0056	0.1172	0.0056	0.0013	0.0056	0.1172	0.6626
					-0.0038	0.0000	0.0000	-0.0005	0.0071	0.7632	0.460	0.0013	0.0056	0.1172	0.0056	0.0013	0.0056	0.1172	0.6626
665	5	7	8	9	1.050	-0.0000	1.06	-0.000	7.2175	-0.0236	6.08	0.04	6.06	0.1382	0.0068	0.04	6.06	0.1382	0.6686
					0.0000	0.0000	0.0000	-0.0005	-0.0345	-0.6141	0.511	0.0014	0.0063	0.1178	0.0063	0.0014	0.0063	0.1178	0.6650
					-0.0042	0.0000	0.0000	-0.0005	-0.0345	-0.6141	0.511	0.0014	0.0063	0.1178	0.0063	0.0014	0.0063	0.1178	0.6650

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key															
665	6	7	1.049	3.16	-0.000	-0.000	0.9580	-0.02	6.11	0.05	6.08	0.1303	0.6500	0.0091	0.0088	0.1137	0.6541
		8	-0.0003	-0.0000	0.0000	0.0005	-1.2609	0.0702	0.0018	0.0015	0.0091						
		9	-0.0035	-0.0000	-0.0005	-0.0005	0.7900	0.0676	0.0015	0.0015	0.0088						
665	7	7	1.050	6.25	-0.000	-0.000	0.7131	-0.03	6.14	0.05	6.11	0.1342	0.6168	0.0112	0.0111	0.1208	0.6076
		8	-0.0013	-0.0001	-0.0000	-0.0003	0.2629	0.0908	0.0024	0.0022	0.0112						
		9	-0.0036	0.0000	-0.0003	-0.0003	0.5333	0.0913	0.0022	0.0022	0.0111						
665	8	7	1.050	9.37	-0.000	-0.000	0.6560	-0.03	6.17	0.04	6.13	0.1287	0.5929	0.0126	0.0126	0.1209	0.5856
		8	-0.0015	-0.0001	-0.0000	-0.0004	0.2088	0.1064	0.0027	0.0026	0.0126						
		9	-0.0045	-0.0000	-0.0004	-0.0004	0.5462	0.1084	0.0026	0.0026	0.0126						
665	9	7	1.050	12.56	-0.000	-0.000	0.5772	-0.03	6.18	0.04	6.15	0.1079	0.5853	0.0143	0.0142	0.1058	0.5737
		8	-0.0016	-0.0001	-0.0000	-0.0006	0.2899	0.1224	0.0026	0.0026	0.0143						
		9	-0.0035	-0.0001	-0.0006	-0.0006	0.8969	0.1240	0.0026	0.0026	0.0142						
665	10	7	1.050	0.00	-0.000	-0.000	-0.02	6.06	0.05	6.04	6.04	0.1484	0.6666	0.0053	0.0050	0.1184	0.6638
		8	0.0000	-0.0001	0.0000	-0.0005	14.7388	0.0403	0.0011	0.0008	0.0053						
		9	-0.0044	0.0000	-0.0005	-0.0005	0.5772	0.0379	0.0008	0.0008	0.0050						
666	1	7	1.050	-3.05	-0.000	-0.000	-0.03	9.09	0.04	9.01	0.1520	0.6797	0.0039	0.0039	0.1013	0.6719	
		8	-0.0011	-0.0001	-0.0000	-0.0005	0.2298	0.0294	0.0008	0.0005	0.0039						
		9	-0.0048	0.0000	-0.0005	-0.0005	0.5901	0.0290	0.0005	0.0005	0.0039						
666	2	7	1.049	-1.02	-0.000	-0.000	-0.02	9.12	0.05	9.03	0.1339	0.6783	0.0068	0.0065	0.1111	0.6727	
		8	-0.0006	-0.0001	-0.0000	-0.0006	0.4078	0.0502	0.0013	0.0010	0.0068						
		9	-0.0038	-0.0000	-0.0006	-0.0006	0.7824	0.0484	0.0010	0.0010	0.0065						
666	3	7	1.050	0.02	-0.000	-0.000	-0.02	9.13	0.05	9.05	0.1285	0.6673	0.0080	0.0076	0.1074	0.6627	
		8	-0.0001	-0.0001	0.0000	-0.0005	2.6384	0.0604	0.0015	0.0012	0.0080						
		9	-0.0037	-0.0000	-0.0005	-0.0005	0.7862	0.0580	0.0012	0.0012	0.0076						
666	4	7	1.052	0.53	-0.000	-0.000	-0.02	9.14	0.05	9.05	0.1285	0.6619	0.0082	0.0082	0.1085	0.6633	
		8	-0.0000	-0.0001	0.0000	-0.0005	7.8851	0.0649	0.0016	0.0013	0.0082						
		9	-0.0038	-0.0000	-0.0005	-0.0005	0.7342	0.0622	0.0013	0.0013	0.0082						
666	5	7	1.051	1.09	-0.000	-0.000	-0.02	9.15	0.05	9.05	0.1268	0.6544	0.0091	0.0088	0.1072	0.6574	
		8	-0.0001	-0.0000	0.0001	-0.0005	3.3158	0.0699	0.0017	0.0014	0.0091						
		9	-0.0042	0.0000	-0.0005	-0.0005	0.5996	0.0672	0.0014	0.0014	0.0088						
666	6	7	1.048	3.18	-0.000	-0.000	-0.02	9.18	0.05	9.08	0.1301	0.6301	0.0107	0.0107	0.1096	0.6352	
		8	-0.0004	-0.0001	0.0000	-0.0005	0.6295	0.0852	0.0022	0.0018	0.0107						
		9	-0.0042	0.0000	-0.0005	-0.0005	0.6004	0.0845	0.0018	0.0018	0.0107						

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key														
666	7	1.050	6.31	-0.000	-0.000	-0.000	0.8290	-0.3652	0.4863	9.20	0.1011	0.0027	0.05	9.11	0.1351	0.6001
	8	-0.0012	-0.0000	-0.0003	-0.0003	-0.1274	0.0000	0.4863	0.1020	0.1011	0.0025	0.0025	0.0025	0.0120	0.1242	0.5918
	9	-0.0038	0.0000													
666	8	1.051	9.37	-0.000	-0.000	0.6628	-0.0047	-0.003	0.4940	9.22	0.1164	0.0027	0.05	9.12	0.1170	0.5872
	8	-0.0015	-0.0001	-0.0004	-0.0004	0.0047	0.0000	0.4940	0.1180	0.1164	0.0026	0.0026	0.0026	0.0136	0.1112	0.5794
	9	-0.0047	-0.0000													
666	9	1.050	12.58	-0.000	-0.000	0.5799	0.0703	-0.003	0.6234	9.23	0.1300	0.0024	0.05	9.13	0.0941	0.5835
	8	-0.0017	-0.0001	-0.0005	-0.0005	0.0703	0.0000	0.6234	0.1313	0.1300	0.0024	0.0024	0.0024	0.0149	0.0937	0.5695
	9	-0.0043	-0.0000													
666	10	1.052	0.003	-0.000	-0.000	-8.6117	0.0448	-0.02	2.8854	9.13	0.0602	0.0015	0.05	9.05	0.1278	0.6650
	8	0.0001	-0.0001	-0.0005	-0.0005	-8.6117	0.0448	-0.02	2.8854	0.0602	0.0015	0.0015	0.0015	0.0076	0.1072	0.6615
	9	-0.0041	-0.0000													
667	1	1.052	-3.01	-0.000	-0.000	1.2443	-0.0519	-0.02	0.4354	12.05	0.0491	0.0012	0.05	12.06	0.1268	0.6840
	8	-0.0009	-0.0002	-0.0005	-0.0005	1.2443	-0.0519	-0.02	0.4354	0.0491	0.0012	0.0012	0.0012	0.0067	0.1040	0.6820
	9	-0.0044	-0.0000													
667	2	1.051	-0.97	-0.000	-0.000	1.0945	-0.0050	-0.02	0.7533	12.08	0.0685	0.0015	0.05	12.08	0.1234	0.6631
	8	-0.0006	-0.0001	-0.0005	-0.0005	1.0945	-0.0050	-0.02	0.7533	0.0685	0.0015	0.0015	0.0015	0.0090	0.1014	0.6613
	9	-0.0037	-0.0000													
667	3	1.050	0.04	-0.000	-0.000	3.1646	-0.0140	-0.02	1.3502	12.09	0.0764	0.0015	0.05	12.09	0.1201	0.6534
	8	-0.0002	-0.0001	-0.0005	-0.0005	3.1646	-0.0140	-0.02	1.3502	0.0764	0.0015	0.0015	0.0015	0.0101	0.1033	0.6527
	9	-0.0039	-0.0000													
667	4	1.049	0.57	-0.000	-0.000	-24.3880	0.0120	-0.02	12.2721	12.10	0.0819	0.0019	0.05	12.09	0.1202	0.6450
	8	0.0000	-0.0001	-0.0005	-0.0005	-24.3880	0.0120	-0.02	12.2721	0.0819	0.0019	0.0019	0.0019	0.0105	0.1025	0.6446
	9	-0.0036	-0.0000													
667	5	1.053	1.10	-0.000	-0.000	-13.7911	0.0377	-0.02	8.2553	12.11	0.0851	0.0017	0.05	12.10	0.1236	0.6373
	8	0.0000	-0.0001	-0.0004	-0.0004	-13.7911	0.0377	-0.02	8.2553	0.0851	0.0017	0.0017	0.0017	0.0108	0.1032	0.6410
	9	-0.0042	-0.0000													
667	6	1.052	3.21	-0.000	-0.000	8.8845	-0.0140	-0.02	0.7859	12.13	0.0962	0.0026	0.05	12.12	0.1376	0.6112
	8	-0.0000	-0.0001	-0.0005	-0.0005	8.8845	-0.0140	-0.02	0.7859	0.0962	0.0026	0.0026	0.0026	0.0114	0.1260	0.6040
	9	-0.0036	-0.0000													
667	7	1.048	6.28	-0.000	-0.000	0.7684	-0.1183	-0.03	0.4368	12.15	0.1115	0.0027	0.05	12.15	0.1283	0.5937
	8	-0.0013	-0.0000	-0.0003	-0.0003	0.7684	-0.1183	-0.03	0.4368	0.1115	0.0027	0.0027	0.0027	0.0132	0.1224	0.5877
	9	-0.0040	-0.0000													

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key															
667	8	7	1.052	9.39	-0.000	0.8959	-0.02	12.16	0.05	12.16	0.0144	0.1049	0.5855					
		6	-0.0011	-0.0002	-0.0004	0.0313	0.2783	0.1236	0.0025	0.0144	0.0144	0.1023	0.5752					
		9	-0.0043	-0.0000	-0.0004	0.0313	0.5221	0.1252	0.0025	0.0144	0.0144	0.1023	0.5752					
667	9	7	1.050	12.60	-0.000	0.5432	-0.03	12.16	0.05	12.16	0.0153	0.0868	0.5806					
		8	-0.0018	-0.0001	-0.0001	0.5432	0.3257	0.1321	0.0022	0.0153	0.0153	0.0868	0.5806					
		9	-0.0045	-0.0000	-0.0005	0.0623	0.5534	0.1340	0.0024	0.0151	0.0151	0.0896	0.5657					
667	10	7	1.050	0.06	-0.000	0.3448	-0.02	12.09	0.05	12.10	0.0101	0.1203	0.6509					
		8	-0.0008	-0.0000	0.0001	0.3448	0.6865	0.0776	0.0018	0.0101	0.0101	0.1203	0.6509					
		9	-0.0039	0.0000	-0.0005	-0.0072	0.6549	0.0764	0.0015	0.0099	0.0099	0.1035	0.6531					
668	1	7	1.050	-3.00	-0.000	0.9948	-0.03	15.10	0.05	15.08	0.0090	0.1201	0.6672					
		8	-0.0010	-0.0002	-0.0000	0.9948	0.3560	0.0681	0.0016	0.0090	0.0090	0.0989	0.6665					
		9	-0.0044	0.0000	-0.0005	-0.0368	0.6299	0.0691	0.0013	0.0092	0.0092	0.0989	0.6665					
668	2	7	1.050	-0.93	-0.000	1.4421	-0.02	15.12	0.05	15.10	0.0110	0.1184	0.6458					
		8	-0.0006	-0.0001	0.0000	1.4421	0.2069	0.0854	0.0020	0.0110	0.0110	0.1184	0.6458					
		9	-0.0038	0.0000	-0.0005	-0.0001	0.6809	0.0860	0.0017	0.0111	0.0111	0.1002	0.6490					
668	3	7	1.049	0.09	-0.000	1.6336	-0.02	15.13	0.05	15.10	0.0116	0.1261	0.6308					
		8	-0.0003	-0.0001	0.0000	1.6336	0.9883	0.0919	0.0023	0.0116	0.0116	0.1261	0.6308					
		9	-0.0036	-0.0000	-0.0005	0.0310	0.7388	0.0935	0.0018	0.0119	0.0119	0.1004	0.6361					
668	4	7	1.050	0.57	-0.000	1.7231	-0.02	15.14	0.05	15.11	0.0117	0.1323	0.6232					
		8	-0.0003	-0.0001	0.0000	1.7231	1.3609	0.0942	0.0024	0.0117	0.0117	0.1323	0.6232					
		9	-0.0038	-0.0000	-0.0005	0.0016	0.6744	0.0952	0.0020	0.0118	0.0118	0.1076	0.6220					
668	5	7	1.047	1.11	-0.000	0.9147	-0.02	15.14	0.05	15.12	0.0120	0.1350	0.6177					
		8	-0.0004	-0.0000	0.0000	0.9147	1.1969	0.0974	0.0026	0.0120	0.0120	0.1350	0.6177					
		9	-0.0040	0.0000	-0.0005	-0.0022	0.6374	0.0968	0.0022	0.0118	0.0118	0.1181	0.6129					
668	6	7	1.051	3.20	-0.000	1.8422	-0.02	15.16	0.05	15.13	0.0129	0.1345	0.6012					
		8	-0.0003	-0.0001	0.0000	1.8422	0.4401	0.1075	0.0028	0.0129	0.0129	0.1345	0.6012					
		9	-0.0043	-0.0000	-0.0005	-0.0388	0.5858	0.1071	0.0026	0.0127	0.0127	0.1217	0.5935					
668	7	7	1.050	6.31	-0.000	0.9030	-0.03	15.17	0.05	15.15	0.0141	0.1139	0.5898					
		8	-0.0011	-0.0002	-0.0003	0.9030	0.3443	0.1201	0.0027	0.0141	0.0141	0.1139	0.5898					
		9	-0.0035	-0.0000	-0.0003	-0.0842	0.5613	0.1218	0.0026	0.0140	0.0140	0.1081	0.5785					
668	8	7	1.048	9.39	-0.000	0.5517	-0.02	15.17	0.04	15.15	0.0152	0.0926	0.5836					
		8	-0.0015	-0.0001	-0.0004	0.5517	0.3003	0.1303	0.0024	0.0152	0.0152	0.0926	0.5836					
		9	-0.0043	-0.0000	-0.0004	0.0484	0.5578	0.1317	0.0024	0.0150	0.0150	0.0923	0.5712					

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																					
663	9	7	1.049	12.60	-0.0000	-0.0000	0.3624	-0.2306	15.17	0.0022	15.15	0.0012	0.5791	0.0040	-0.0000	-0.0000	0.0498	0.6051	0.1381	0.0023	0.0158	0.0863	0.5644
668	10	7	1.051	0.05	-0.0000	-0.0000	1.4641	-0.8989	15.14	0.0022	15.10	0.1251	0.6274	-0.0004	-0.0000	-0.0000	-0.0243	0.6123	0.0930	0.0018	0.0114	0.1007	0.6348
669	5	7	1.048	-3.11	-0.0000	-0.0000	0.2830	-3.01	2.97	2.95	-3.01	0.0820	0.6075	-0.0127	-0.0016	-0.0015	0.2313	0.5508	-0.0411	-0.0014	-0.0009	0.1751	0.6616
669	6	7	1.045	-0.02	-0.0000	-0.0000	0.2860	-3.00	3.01	2.95	-2.98	0.2191	0.6796	-0.0145	-0.0014	-0.0015	0.2315	0.5696	-0.0115	-0.0006	-0.0015	0.2975	0.6644
669	7	7	1.048	0.51	-0.0000	-0.0000	0.3075	-3.00	3.02	2.95	-2.97	0.1987	0.6725	-0.0109	-0.0013	-0.0010	0.3075	0.6385	0.0228	0.0009	0.0030	0.3869	0.6093
669	8	7	1.043	-0.01	-0.0000	-0.0000	0.3073	-3.01	3.01	2.95	-2.97	0.2156	0.6776	-0.0145	-0.0014	-0.0015	0.2420	0.5765	-0.0117	-0.0007	-0.0015	0.2994	0.6726
669	9	7	1.043	1.01	-0.0000	-0.0000	0.2656	-3.00	3.03	2.95	-2.96	0.1887	0.6729	-0.0112	-0.0013	-0.0005	0.2485	0.6069	0.0280	0.0010	0.0005	0.6413	0.7113
669	10	7	1.050	2.07	-0.0000	-0.0000	0.2768	-3.00	3.04	2.95	-2.95	0.1678	0.6696	-0.0111	-0.0014	-0.0002	0.2646	0.5822	0.0024	0.0001	0.0002	-0.2443	0.6167
669	11	7	1.050	3.11	-0.0000	-0.0000	0.2917	-3.00	3.06	2.95	-2.93	0.1554	0.6707	-0.0112	-0.0014	-0.0013	0.2882	0.5817	0.0104	0.0001	0.0013	0.0815	0.6343
669	12	7	1.048	4.15	-0.0000	-0.0000	0.3003	-3.01	3.07	2.95	-2.92	0.1464	0.6661	-0.0119	-0.0016	-0.0024	0.3003	0.6691	0.0191	0.0005	0.0024	0.1375	0.6387
669	13	7	1.049	-0.04	-0.0000	-0.0000	0.2568	-3.01	3.01	2.95	-2.98	0.2080	0.6694	-0.0139	-0.0017	-0.0016	0.2568	0.6099	-0.0119	-0.0006	-0.0016	0.2928	0.6857

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																				
670	3	7	1.047	-3.10	-0.000	-0.000	0.2737	-3.02	-0.10	2.94	-0.06	0.1408	0.6569	0.0144	-0.0007	-0.0018	0.2737	0.6490	-0.0263	2.94	-0.034	0.2364	0.6529
670	4	7	1.047	-0.04	-0.000	-0.000	0.2618	-3.02	-0.06	2.95	-0.03	0.7664	0.4651	0.0143	-0.0007	-0.0018	0.2618	0.6112	0.0011	2.95	0.0001	-0.2456	0.6364
670	5	7	1.049	0.47	-0.000	-0.000	0.2723	-3.02	-0.05	2.94	-0.03	0.3331	0.6190	0.0119	-0.0006	-0.0019	0.2723	0.6276	0.0061	2.94	0.0006	0.0151	0.6632
670	6	7	1.047	1.00	-0.000	-0.000	0.2725	-3.02	-0.04	2.94	-0.01	0.2584	0.6328	0.0117	-0.0006	-0.0019	0.2725	0.6362	0.0094	2.94	0.0011	0.0616	0.6394
670	7	7	1.049	2.07	-0.000	-0.000	0.2473	-3.02	-0.02	2.94	-0.00	0.2172	0.6398	0.0118	-0.0008	-0.0020	0.2473	0.6265	0.0185	2.94	0.0023	0.1136	0.6219
670	8	7	1.048	3.12	-0.000	-0.000	0.2413	-3.02	-0.01	2.95	-0.00	0.1900	0.6387	0.0124	-0.0009	-0.0020	0.2413	0.6309	0.0284	2.95	0.0036	0.1329	0.6420
670	9	7	1.048	4.18	-0.000	-0.000	0.2207	-3.02	0.00	2.95	0.02	0.1724	0.6466	0.0132	-0.0010	-0.0021	0.2207	0.6141	0.381	2.95	0.0049	0.1323	0.6497
670	10	7	1.045	-0.03	-0.000	-0.000	0.2541	-3.02	-0.06	2.94	-0.03	0.5329	0.3039	0.0127	-0.0006	-0.0018	0.2541	0.6018	0.0029	2.94	0.0003	-0.2626	0.5733
671	1	7	1.051	-3.10	-0.000	-0.000	0.0277	0.49	-0.10	-0.050	-0.06	0.1570	0.6715	0.0039	0.0000	-0.0005	0.0277	0.6866	-0.0243	-0.0007	-0.0030	0.2264	0.6576
671	2	7	1.050	-0.02	-0.000	-0.000	0.1308	0.49	-0.06	-0.048	-0.03	0.4258	0.5821	0.0048	0.0001	-0.0004	0.1308	0.7459	0.0019	-0.0001	0.0002	-0.5581	0.6118
671	3	7	1.050	0.49	-0.000	-0.000	0.1511	0.49	-0.05	-0.048	-0.02	0.2424	0.6113	0.0051	0.0001	-0.0004	0.1511	0.7160	0.0066	-0.0003	0.0006	0.2424	0.6113

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																				
671	4	7	1.047	0.99	-0.007	0.1246	0.6623	0.49	-0.05	-0.48	-0.013	0.2241	0.6378	0.0054	0.0001	-0.0005	0.049	0.0106	0.0004	0.0010	0.0721	0.6198
671	5	7	1.049	2.06	-0.007	0.1539	0.6936	0.49	-0.02	-0.48	-0.004	0.1924	0.6265	0.0052	0.0001	-0.0005	0.6936	0.0198	0.0007	0.0024	0.1192	0.6134
671	6	7	1.044	3.12	-0.006	0.0257	0.5686	0.49	-0.01	-0.49	0.000	0.1774	0.6421	0.0054	0.0000	-0.0004	0.5686	0.0303	0.0010	0.0039	0.1400	0.6358
671	7	7	1.048	4.14	-0.005	-0.0098	0.5514	0.49	0.00	-0.48	0.000	0.1547	0.6508	0.0048	-0.0000	-0.0004	0.5514	0.0395	0.0013	0.0051	0.1359	0.6458
671	8	7	1.049	-0.02	-0.007	0.1718	0.8061	0.49	-0.06	-0.48	0.000	0.3299	0.5324	0.0046	0.0001	-0.0005	0.8061	0.0024	0.0001	0.0002	0.4396	0.5287
672	1	7	1.048	3.09	-0.005	0.0916	0.7628	0.49	-0.55	-0.48	0.000	0.1544	0.6651	0.0034	0.0000	-0.0005	0.7628	0.0272	-0.0008	-0.0036	0.2376	0.6665
672	2	7	1.045	-0.02	-0.007	0.1379	0.7433	0.48	-0.50	-0.48	0.000	0.6827	0.3253	0.0045	0.0001	-0.0004	0.7433	0.0004	0.0000	0.0000	0.0752	0.6164
672	3	7	1.050	0.47	-0.006	0.0678	0.6245	0.48	-0.50	-0.48	0.000	0.2813	0.5801	0.0055	0.0000	-0.0004	0.6245	0.0040	0.0002	0.0004	0.0608	0.6428
672	4	7	1.049	1.03	-0.007	0.1353	0.6510	0.49	-0.49	-0.48	0.000	0.2888	0.6146	0.0055	0.0000	-0.0004	0.6510	0.0082	0.0003	0.0010	0.0949	0.6422
672	5	7	1.048	2.06	-0.007	0.1274	0.6559	0.49	-0.47	-0.47	0.000	0.1966	0.6243	0.0053	0.0001	-0.0004	0.6559	0.0172	0.0006	0.0021	0.1276	0.6203
672	6	7	1.048	3.13	-0.006	0.0969	0.6394	0.49	-0.46	-0.48	0.000	0.1794	0.6342	0.0051	0.0000	-0.0004	0.6394	0.0272	0.0009	0.0034	0.1379	0.6404

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																
672	7	1.049	4.15	-0.00	0.0070	0.49	-0.44	-0.48	0.52	0.1686	0.6478	0.0048	0.0005	0.0012	0.0047	0.0052	0.1327	0.6460
	8	0.0048	0.0000	0.0004	-0.1860	0.5392	0.0363	0.0010	0.0052									
	9	-0.0020	0.0000	-0.0004	-0.1860	1.1033	0.0408	0.0010	0.0052									
672	7	1.048	-0.03	-0.00	0.1273	0.48	-0.50	-0.48	0.47	0.3347	-0.5006	0.0048	0.0007	0.0000	0.0004	0.0004	-0.0898	-0.5790
	8	0.0048	0.0000	0.0005	-0.1040	0.7209	0.0004	0.0000	-0.0000									
	9	-0.0028	0.0000	-0.0005	-0.1040	0.8897	0.0038	-0.0000	0.0004									
673	1	1.046	-3.11	-0.00	0.1287	1.02	-0.10	-1.03	0.06	0.1676	0.6621	0.0072	0.0009	0.0007	-0.0031	0.0030	0.2262	0.6727
	8	0.0072	0.0001	0.0009	0.0462	0.6727	-0.0228	-0.0010	-0.0030									
	9	-0.0057	-0.0000	-0.0009	0.0462	0.8225	-0.0228	-0.0010	-0.0030									
673	2	1.044	-0.03	-0.00	0.1510	1.02	-0.05	-1.03	0.02	0.2755	0.5458	0.0083	0.0011	0.0001	0.0003	0.0001	-0.5435	0.5740
	8	0.0083	0.0002	0.0011	0.0518	0.6865	0.0028	0.0001	0.0001									
	9	-0.0053	-0.0000	-0.0008	0.0518	0.8216	0.0015	-0.0001	0.0001									
673	3	1.049	0.47	-0.00	0.1071	1.02	-0.04	-1.03	0.02	0.2225	0.6316	0.0091	0.0011	0.0003	0.0005	0.0005	0.0011	0.6318
	8	0.0091	0.0001	0.0008	0.0072	0.6155	0.0069	0.0000	0.0003									
	9	-0.0056	-0.0000	-0.0008	0.0072	0.7083	0.0046	0.0000	0.0005									
673	4	1.045	1.02	-0.00	0.1570	1.02	-0.04	-1.03	0.02	0.2109	0.6385	0.0088	0.0011	0.0004	0.0010	0.0010	0.0567	0.5962
	8	0.0088	0.0002	0.0011	0.0603	0.6555	0.0112	0.0000	0.0010									
	9	-0.0050	-0.0000	-0.0009	0.0603	0.8869	0.0088	0.0000	0.0010									
673	5	1.047	2.05	-0.00	0.1433	1.02	-0.02	-1.03	0.00	0.1856	0.6273	0.0088	0.0011	0.0007	0.0020	0.0020	0.1136	0.5987
	8	0.0088	0.0002	0.0008	0.0545	0.6381	0.0206	0.0003	0.0020									
	9	-0.0052	-0.0000	-0.0008	0.0545	0.8580	0.0171	0.0003	0.0020									
673	6	1.045	3.09	-0.00	0.0987	1.02	-0.00	-1.02	0.01	0.1713	0.6432	0.0086	0.0010	0.0007	0.0033	0.0033	0.1337	0.6229
	8	0.0086	0.0001	0.0010	0.0339	0.5879	0.0305	0.0000	0.0033									
	9	-0.0053	-0.0000	-0.0008	0.0339	0.7685	0.0266	0.0000	0.0033									
673	7	1.048	4.15	-0.00	0.0918	1.02	0.00	-1.02	0.02	0.1591	0.6480	0.0077	0.0009	0.0012	0.0047	0.0047	0.1364	0.6374
	8	0.0077	0.0001	0.0009	-0.0091	0.6049	0.0398	0.0010	0.0047									
	9	-0.0051	0.0000	-0.0007	-0.0091	0.7269	0.0372	0.0010	0.0047									
673	8	1.050	-0.03	-0.00	0.0822	1.02	-0.06	-1.02	0.03	0.2372	0.5000	0.0089	0.0010	0.0001	0.0002	0.0001	-0.5982	0.3424
	8	0.0089	0.0001	0.0010	0.0265	0.5829	0.0027	0.0001	0.0002									
	9	-0.0059	0.0000	-0.0007	-0.0265	0.6547	0.0015	-0.0001	0.0001									
674	1	1.048	-3.09	-0.00	0.1642	1.02	-1.00	-1.02	0.93	0.1575	0.6684	0.0062	0.0008	0.0009	0.0039	0.0022	0.2658	0.6932
	8	0.0062	0.0002	0.0008	0.0202	0.7048	-0.0297	-0.0008	-0.0022									
	9	-0.0063	0.0000	-0.0008	-0.0202	0.6430	-0.0159	-0.0008	-0.0022									

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key															
674	2	7	1.050	-0.02	-0.00	0.1233	0.6217	-0.95	-1.02	-0.0003	0.2279	1.0965						
		8	0.0084	0.0002	0.0010	0.0081	0.7872	-0.0014	-0.0000	-0.0007	0.0175	0.6322						
		9	-0.0051	-0.0000	-0.0008	0.0001	0.0061	0.0061	0.0000	0.0007	0.0000	0.6322						
674	3	7	1.047	0.47	-0.00	0.1394	0.6409	-0.94	-1.02	0.96	0.3069	0.4939						
		8	0.0086	0.0002	0.0011	0.0204	0.6937	0.0222	0.0001	0.0012	0.0758	0.6348						
		9	-0.0054	0.0000	-0.0007	-0.0004	0.0102	0.0102	0.0001	0.0012	0.0000	0.6348						
674	4	7	1.049	1.02	-0.00	0.1160	0.6034	-0.94	-1.02	0.97	0.2189	0.5870						
		8	0.0088	0.0002	0.0010	0.0043	0.7120	0.0066	0.0003	0.0017	0.1064	0.6217						
		9	-0.0054	0.0000	-0.0007	-0.0004	0.0143	0.0143	0.0003	0.0017	0.0000	0.6217						
674	5	7	1.048	2.05	-0.00	0.1240	0.6095	-0.92	-1.01	0.99	0.1923	0.6161						
		8	0.0087	0.0002	0.0010	0.0243	0.6682	0.0149	0.0005	0.0028	0.1288	0.6187						
		9	-0.0054	0.0000	-0.0007	-0.0004	0.0230	0.0230	0.0005	0.0028	0.0000	0.6187						
674	6	7	1.046	3.10	-0.00	0.1635	0.6663	-0.90	-1.01	1.00	0.1817	0.6293						
		8	0.0077	0.0002	0.0010	0.0303	0.8575	0.0242	0.0008	0.0042	0.1296	0.6327						
		9	-0.0045	-0.0000	-0.0007	0.0003	0.0535	0.0335	0.0008	0.0042	0.0000	0.6327						
674	7	7	1.049	4.15	-0.00	0.1210	0.6271	-0.89	-1.01	1.01	0.1672	0.6387						
		8	0.0073	0.0001	0.0009	0.0172	0.8723	0.0337	0.0011	0.0043	0.1279	0.6436						
		9	-0.0042	-0.0000	-0.0007	0.0003	0.0518	0.0443	0.0011	0.0057	0.0000	0.6436						
674	8	7	1.047	-0.02	-0.00	0.1067	0.6167	-0.95	-1.02	0.96	0.2560	1.1436						
		8	0.0083	0.0001	0.0010	0.0142	0.8518	0.0012	0.0000	0.0007	0.0161	0.5948						
		9	-0.0049	-0.0000	-0.0008	0.0003	0.0518	0.0064	0.0000	0.0007	0.0000	0.5948						
675	6	7	1.051	-3.09	0.00	0.1594	0.6711	-0.10	-2.12	-0.06	0.1826	0.6741						
		8	0.0132	0.0004	0.0017	0.0538	0.6192	0.0227	0.0008	0.0030	0.2168	0.6699						
		9	-0.0129	-0.0001	-0.0016	0.0003	0.6592	0.0234	0.0010	0.0031	0.0000	0.6699						
675	7	7	1.050	-0.03	-0.00	0.1041	0.6170	-0.06	-2.11	-0.02	0.2031	0.6164						
		8	0.0147	0.0003	0.0018	0.0634	0.6592	0.0026	0.0001	0.0000	-1.0585	0.5566						
		9	-0.0121	-0.0001	-0.0016	0.0003	0.6592	0.0008	0.0001	0.0000	0.0000	0.5566						
675	8	7	1.053	0.47	-0.00	0.1318	0.6490	-0.05	-2.12	-0.01	0.1814	0.6243						
		8	0.0145	0.0003	0.0018	0.0321	0.5691	0.0071	0.0002	0.0005	-0.0129	0.6713						
		9	-0.0131	-0.0000	-0.0014	0.0003	0.5691	0.0039	0.0000	0.0005	0.0000	0.6713						
675	9	7	1.051	1.03	-0.00	0.1731	0.6918	-0.05	-2.11	-0.01	0.1803	0.6274						
		8	0.0138	0.0004	0.0019	0.0824	0.6969	0.0112	0.0004	0.0009	0.0866	0.6508						
		9	-0.0119	-0.0001	-0.0016	0.0003	0.6969	0.0075	0.0001	0.0009	0.0000	0.6508						

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																			
675	10	1.050	2.06	-0.00	0.1226	2.08	-0.02	-2.12	0.00	0.1705	0.6267	0.0146	0.0003	0.0018	0.0493	0.6276	0.0206	0.0007	0.0025	0.1208	0.6158
		-0.0130	-0.0001	-0.0015		0.5865	0.0163	0.0003	0.0020			-0.0130	-0.0015	-0.0015							
675	11	1.052	3.14	-0.00	0.1150	2.07	-0.01	-2.12	0.01	0.1643	0.6418	0.0140	0.0003	0.0017	0.0592	0.6216	0.0303	0.0009	0.0038	0.1425	0.6348
		-0.0126	-0.0001	-0.0015		0.6174	0.0258	0.0007	0.0032			-0.0126	-0.0015	-0.0015							
675	12	1.053	4.16	-0.00	0.1191	2.07	0.00	-2.12	0.02	0.1546	0.6502	0.0133	0.0003	0.0016	0.0502	0.6300	0.0400	0.0012	0.0052	0.1412	0.6502
		-0.0125	-0.0001	-0.0014		0.5844	0.0365	0.0010	0.0047			-0.0125	-0.0014	-0.0014							
675	13	1.054	0.01	-0.00	0.1224	2.07	-0.06	-2.12	0.02	0.1415	0.5882	0.0144	0.0003	0.0018	0.0415	0.6327	0.0030	0.0002	0.0003	0.1415	0.5882
		-0.0124	-0.0001	-0.0015		0.6332	0.0008	0.0002	0.0000			-0.0124	-0.0015	-0.0015							
676	1	1.052	3.11	-0.00	0.1601	2.07	-1.89	-2.11	1.93	0.1656	0.6842	0.0122	0.0003	0.0016	0.0621	0.6589	-0.0346	-0.0011	-0.0047	0.1656	0.6842
		-0.0118	-0.0001	-0.0015		0.6683	-0.0113	-0.0006	0.0015			-0.0118	-0.0015	-0.0015							
676	2	1.051	0.01	-0.00	0.1307	2.08	-0.05	-2.11	1.97	0.2334	0.7637	0.0136	0.0003	0.0017	0.0611	0.6776	-0.0105	-0.0001	-0.0013	0.2334	0.7637
		-0.0113	-0.0001	-0.0015		0.6366	-0.0105	-0.0001	0.0013			-0.0113	-0.0015	-0.0015							
676	3	1.052	0.48	-0.00	0.1482	2.07	-1.84	-2.11	1.98	0.4206	1.0880	0.0136	0.0004	0.0017	0.0420	0.6561	-0.0149	-0.0003	-0.0018	0.4206	1.0880
		-0.0117	-0.0000	-0.0014		0.6348	-0.0149	-0.0003	0.0018			-0.0117	-0.0014	-0.0014							
676	4	1.053	1.00	-0.00	0.1281	2.07	-1.83	-2.11	1.98	0.2408	0.5599	0.0142	0.0003	0.0017	0.0582	0.6285	0.0014	0.0004	0.0023	0.2408	0.5599
		-0.0113	-0.0001	-0.0015		0.6699	0.0187	0.0004	0.0023			-0.0113	-0.0015	-0.0015							
676	5	1.053	2.07	-0.00	0.1286	2.08	-1.82	-2.11	2.00	0.1845	0.6078	0.0141	0.0003	0.0017	0.0515	0.6146	0.0101	0.0007	0.0036	0.1845	0.6078
		-0.0114	-0.0001	-0.0014		0.6379	0.0287	0.0007	0.0036			-0.0114	-0.0014	-0.0014							
676	6	1.051	3.13	-0.00	0.1607	2.07	-1.80	-2.11	2.01	0.1769	0.6161	0.0131	0.0004	0.0017	0.0588	0.6536	0.0191	0.0006	0.0023	0.1769	0.6161
		-0.0113	-0.0001	-0.0014		0.6308	0.0392	0.0010	0.0023			-0.0113	-0.0014	-0.0014							
676	7	1.053	4.16	-0.00	0.1147	2.08	-1.79	-2.11	2.02	0.1709	0.6303	0.0132	0.0003	0.0016	0.0593	0.6107	0.0282	0.0009	0.0035	0.1709	0.6303
		-0.0108	-0.0001	-0.0013		0.6144	0.0498	0.0013	0.0035			-0.0108	-0.0013	-0.0013							

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key														
676	8	7	1.050	-0.004	-0.003	-0.0017	0.1774	0.6884	2.07	-1.84	-0.0056	-2.11	1.97	0.2623	0.6000	
		9	0.0130	0.0004	0.0004	0.0017	0.0651	0.7476	0.6884	0.0106	0.0106	0.0002	-0.0008	0.0820	0.6138	
		9	-0.0108	-0.0001	-0.0001	-0.0016							0.0013			
677	1	7	1.054	-3.10	-0.0005	-0.0044	0.1558	0.6899	5.07	-0.09	-0.0211	-5.05	-0.07	0.2073	0.6661	
		8	0.0322	0.0010	0.0010	0.0044	0.0915	0.6566	0.6899	-0.0245	-0.0245	-0.0010	-0.0028	0.2134	0.6819	
		9	-0.0312	-0.0005	-0.0005	-0.0041							-0.0033			
677	2	7	1.054	-0.003	-0.0007	-0.0040	0.1149	0.6512	5.07	-0.05	-0.0039	-5.05	-0.03	0.0604	0.6122	
		8	0.0345	0.0007	0.0007	0.0044	0.0939	0.6253	0.6512	0.0006	0.0006	0.0002	0.0004	1.6385	0.6555	
		9	-0.0316	-0.0005	-0.0005	-0.0039							-0.0001			
677	3	7	1.049	0.48	0.0007	-0.0045	0.1281	0.6678	5.07	-0.05	-0.0027	-5.05	-0.02	0.1311	0.6282	
		8	0.0342	0.0008	0.0008	0.0045	0.1224	0.6864	0.6678	0.0087	0.0087	0.0002	0.0011	-0.0249	0.5488	
		9	-0.0304	-0.0007	-0.0007	-0.0041							0.0003			
677	4	7	1.049	1.01	0.0007	-0.0045	0.1288	0.6717	5.07	-0.04	-0.0126	-5.05	-0.01	0.1443	0.6252	
		8	0.0339	0.0008	0.0008	0.0045	0.1207	0.6757	0.6717	0.0061	0.0061	0.0001	0.0015	0.0838	0.5839	
		9	-0.0308	-0.0007	-0.0007	-0.0041							0.0007			
677	5	7	1.051	2.06	0.0007	-0.0045	0.1217	0.6662	5.07	-0.02	-0.0219	-5.05	-0.00	0.1498	0.6241	
		8	0.0337	0.0008	0.0008	0.0045	0.1153	0.6570	0.6662	0.0146	0.0146	0.0004	0.0027	0.1375	0.6059	
		9	-0.0312	-0.0007	-0.0007	-0.0041							0.0017			
677	6	7	1.051	3.12	0.0007	-0.0043	0.1213	0.6701	5.07	-0.00	-0.0324	-5.05	0.00	0.1490	0.6409	
		8	0.0326	0.0007	0.0007	0.0043	0.1232	0.6642	0.6701	0.0249	0.0249	0.0007	0.0041	0.1501	0.6174	
		9	-0.0313	-0.0007	-0.0007	-0.0041							0.0030			
677	7	7	1.050	4.15	0.0008	-0.0041	0.1060	0.6585	5.07	0.00	0.0414	-5.05	0.02	0.1426	0.6482	
		8	0.0318	0.0008	0.0008	0.0041	0.1448	0.7046	0.6585	0.0350	0.0350	0.0010	0.0044	0.1448	0.6342	
		9	-0.0296	-0.0008	-0.0008	-0.0041							0.0044			
677	8	7	1.049	-0.02	-0.0006	-0.0040	0.1211	0.6599	5.07	-0.05	-0.0043	-5.05	-0.03	0.0411	0.5884	
		8	0.0342	0.0008	0.0008	0.0040	0.1029	0.6586	0.6599	0.0043	0.0043	0.0001	0.0005	2.2780	1.5472	
		9	-0.0310	-0.0006	-0.0006	-0.0040							-0.0001			
678	1	7	1.051	-3.10	0.0009	-0.0039	0.1521	0.6577	5.07	-5.12	-0.0015	-5.05	5.01	0.1492	0.6870	
		8	0.0297	0.0009	0.0009	0.0039	0.0952	0.6380	0.6577	-0.0531	-0.0531	-0.0004	-0.0073	-0.1461	0.6951	
		9	-0.0301	-0.0009	-0.0009	-0.0038							0.0004			
678	2	7	1.053	-0.02	-0.0006	-0.0042	0.1362	0.6684	5.07	-5.07	-0.0233	-5.05	5.03	0.1838	0.6963	
		8	0.0316	0.0006	0.0006	0.0042	0.1156	0.6679	0.6684	-0.0280	-0.0280	-0.0007	-0.0032	0.1269	0.6485	
		9	-0.0283	-0.0006	-0.0006	-0.0037							0.0036			

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key															
678	3	7 8 9	1.052 0.0320 -0.0287	0.50 0.0008 -0.0006	-0.00 0.0042 -0.0037	0.1365 0.1125	5.07 0.6671 0.6453	-5.06 -0.0175 0.0333	-5.05 -0.0007 0.0008	5.05 -0.0025 0.0043	0.2007 0.1289	0.7131 0.6544						
67A	4	7 8 9	1.047 0.0332 -0.0288	1.00 0.0007 -0.0006	-0.00 0.0042 -0.0037	0.1150 0.1114	5.07 0.6405 0.6416	-5.05 -0.0131 0.0378	-5.05 -0.0005 0.0009	5.06 -0.0019 0.0049	0.2105 0.1308	0.7259 0.6595						
67A	5	7 8 9	1.050 0.0322 -0.0277	2.09 0.0008 -0.0006	-0.00 0.0042 -0.0036	0.1358 0.1246	5.07 0.6632 0.6652	-5.04 -0.0044 0.0476	-5.05 -0.0002 0.0012	5.07 -0.0007 0.0062	0.3019 0.1288	0.8259 0.6604						
678	6	7 8 9	1.047 0.0318 -0.0279	3.12 0.0008 -0.0006	-0.00 0.0042 -0.0035	0.1341 0.1194	5.07 0.6628 0.6357	-5.03 0.0025 0.0578	-5.05 0.0000 0.0014	5.06 0.0002 0.0076	0.1538 0.1271	0.4525 0.6579						
67A	7	7 8 9	1.050 0.0311 -0.0263	4.16 0.0007 -0.0007	-0.00 0.0040 -0.0035	0.1192 0.1387	5.07 0.6510 0.6812	-5.01 0.0102 0.0678	-5.05 0.0003 0.0016	5.10 0.0011 0.0088	0.1536 0.1225	0.5644 0.6496						
678	8	7 8 9	1.049 0.0326 -0.0291	-0.01 0.0007 -0.0005	-0.00 0.0042 -0.0037	0.1187 0.1001	5.07 0.6438 0.6351	-5.07 -0.0230 0.0282	-5.04 -0.0008 0.0007	5.04 -0.0032 0.0036	0.1870 0.1292	0.6971 0.6491						
679	3	7 8 9	1.051 0.0018 -0.0007	-3.15 0.0001 -0.0001	0.00 0.0002 -0.0002	-0.3900 -0.8746	-0.02 0.6579 1.2865	-3.03 -0.0453 -0.0435	0.05 -0.0012 -0.0015	-3.03 -0.0061 -0.0058	0.1366 0.1733	0.6739 0.6745						
679	4	7 8 9	1.050 0.0016 -0.0003	-0.07 0.0000 0.0001	0.00 0.0003 -0.0001	0.1529 -2.2261	-0.01 1.1847 2.3105	-2.97 -0.0144 -0.0139	0.05 -0.0004 -0.0007	-2.98 -0.0020 -0.0019	0.1431 0.2771	0.6919 0.7051						
679	5	7 8 9	1.052 0.0017 -0.0005	0.00 0.0000 0.0001	-0.00 0.0003 -0.0001	0.0883 -1.5877	-0.02 0.6864 1.4194	-2.97 -0.0138 -0.0134	0.05 -0.0004 -0.0007	-2.99 -0.0019 -0.0018	0.1454 0.2756	0.6980 0.6920						
679	6	7 8 9	1.050 0.0024 -0.0001	0.98 0.0000 0.0001	-0.00 0.0003 -0.0001	0.0010 -5.8799	-0.01 0.7644 6.9390	-2.96 -0.0053 -0.0056	0.05 -0.0001 -0.0005	-2.98 -0.0007 -0.0008	0.1033 0.4583	0.7439 0.7292						
679	7	7 8 9	1.048 0.0027 -0.0004	2.03 0.0000 -0.0001	-0.00 0.0003 -0.0001	-0.0130 -2.0074	-0.02 0.6587 1.5915	-2.95 0.0023 0.0009	0.05 -0.0002 -0.0001	-2.96 0.0000 0.0000	0.4661 -0.8424	0.4940 0.2311						

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																			
679	8	1.047	3.09	0.00	-0.1464	-0.02	-2.93	0.05	-2.95	0.2403	0.6174	0.0030	-0.0001	0.0003	0.5110	0.0107	0.0005	-2.95	0.0013	0.0796	0.6057
	8	0.0001	0.0000	0.0003	5.0635	0.7052	0.0089	0.0001	0.0010												
679	9	1.047	4.12	-0.00	-0.0298	-0.02	-2.92	0.06	-2.94	0.2052	0.6140	0.0020	-0.0001	0.0003	0.7879	0.0190	0.0007	-2.94	0.0023	0.1266	0.5949
	8	0.0012	0.0000	0.0003	0.7640	0.4595	0.0180	0.0004	0.0021												
679	10	1.049	-0.05	-0.00	-0.0688	-0.02	-2.98	0.05	-2.99	0.1577	0.7168	0.0020	-0.0001	0.0003	0.8111	0.0140	0.0004	-2.99	0.0020	0.2767	0.7136
	8	0.0003	0.0000	0.0003	2.1943	0.0007	0.0138	0.0007	0.0019												
680	1	1.050	-3.11	-0.00	-0.3017	-0.02	-0.99	0.06	-1.04	0.1521	0.6668	0.0016	-0.0001	0.0002	0.7441	0.0309	0.0009	-1.04	0.0041	0.2003	0.6733
	8	0.0008	0.0001	0.0001	1.0300	0.9665	0.0300	0.0012	0.0040												
680	2	1.050	-0.03	-0.00	-0.0585	-0.02	-0.95	0.05	-1.00	0.1319	0.8214	0.0020	-0.0001	0.0003	0.8692	0.0027	0.0000	-1.00	0.0004	0.6538	0.6717
	8	0.0005	0.0000	0.0003	1.6132	1.7097	0.0030	0.0004	0.0004												
680	3	1.049	0.02	-0.00	-0.2071	-0.02	-0.95	0.05	-1.00	0.1273	0.8814	0.0026	-0.0001	0.0002	0.5596	0.0021	0.0000	-1.00	0.0003	0.7318	0.6999
	8	0.0004	0.0001	0.0001	1.8738	2.1588	0.0026	0.0003	0.0003												
680	4	1.048	1.00	-0.00	-0.0237	-0.01	-0.94	0.05	-0.99	0.2629	0.6067	0.0025	-0.0001	0.0003	0.7332	0.0051	0.0002	-0.99	0.0006	0.1187	0.5984
	8	0.0000	0.0000	0.0003	2.5237	3.1404	0.0030	0.0000	0.0003												
680	5	1.050	2.05	-0.00	-0.0259	-0.01	-0.92	0.05	-0.98	0.2072	0.6241	0.0027	-0.0002	0.0003	0.6401	0.0134	0.0005	-0.98	0.0016	0.0960	0.6217
	8	0.0008	0.0002	0.0000	1.3984	0.4413	0.0110	0.0002	0.0013												
680	6	1.049	3.09	-0.00	-0.0911	-0.02	-0.91	0.06	-0.96	0.1876	0.6301	0.0028	-0.0001	0.0002	0.5792	0.0229	0.0008	-0.96	0.0028	0.1290	0.6135
	8	0.0014	0.0001	0.0003	0.4702	0.7894	0.0204	0.0005	0.0025												
680	7	1.045	4.15	-0.00	-0.1944	-0.02	-0.89	0.06	-0.95	0.1699	0.6360	0.0022	-0.0002	0.0000	0.6328	0.0328	0.0011	-0.95	0.0041	0.1376	0.6328
	8	0.0006	0.0002	0.0000	1.6414	0.6403	0.0309	0.0008	0.0039												
680	8	1.050	-2.12	-0.00	-0.2542	-0.02	-0.98	0.06	-1.03	0.1624	0.6735	0.0013	-0.0002	0.0002	0.9122	0.0213	0.0006	-1.03	0.0028	0.2358	0.6902
	8	0.0008	0.0002	0.0001	1.2927	0.9378	0.0203	0.0009	0.0028												

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																					
680	9	1.048	-0.003	-0.003	0.0529	-0.02	0.9794	-0.0025	0.05	-1.00	0.1774	0.0646	0.0019	0.0000	0.0003	0.0529	-0.02	0.9794	-0.0025	0.05	-1.00	0.1774	0.0646
	8	-0.0004	0.0001	0.0000	-2.0725	1.8374	-0.0032	-0.0004	-0.0004	-0.0004	0.6447	0.7015	-0.0004	0.0000	-0.0001	-0.0004	0.6447	1.8374	-0.0032	-0.0004	-0.0004	0.6447	0.7015
681	1	1.047	-3.12	0.0000	-0.3094	-0.03	0.9682	-0.0248	0.05	-0.06	0.1621	0.6618	0.0011	0.0000	0.0002	-0.3094	-0.03	0.9682	-0.0248	0.05	-0.06	0.1621	0.6618
	8	-0.0010	0.0002	0.0000	-1.0111	0.8468	-0.0231	-0.0010	-0.0010	-0.0030	0.2269	0.6675	-0.0010	0.0000	-0.0001	-1.0111	0.8468	-0.0231	-0.0010	-0.0010	-0.0030	0.2269	0.6675
681	2	1.048	-0.02	0.0000	-0.1395	-0.03	0.7873	-0.0018	0.05	-0.02	0.3439	0.6150	0.0021	0.0000	0.0003	-0.1395	-0.03	0.7873	-0.0018	0.05	-0.02	0.3439	0.6150
	8	0.0000	0.0001	0.0000	1.687621	0.0000	0.0000	0.0000	-0.0001	0.0001	-0.6315	0.6599	0.0000	0.0000	-0.0002	1.687621	0.0000	0.0000	0.0000	-0.0001	0.0001	-0.6315	0.6599
681	3	1.048	0.02	0.0000	0.0250	-0.03	1.0383	-0.0022	0.06	-0.03	0.3005	0.5809	0.0018	0.0000	0.0003	0.0250	-0.03	1.0383	-0.0022	0.06	-0.03	0.3005	0.5809
	8	-0.0007	0.0002	0.0000	-1.3551	0.8507	0.0015	-0.0001	-0.0001	0.0001	-0.5109	0.6337	-0.0007	0.0000	-0.0001	-1.3551	0.8507	0.0015	-0.0001	-0.0001	0.0001	-0.5109	0.6337
681	4	1.048	1.01	0.0000	-0.0056	-0.03	0.8529	-0.0100	0.05	-0.01	0.2259	0.6485	0.0021	0.0000	0.0003	-0.0056	-0.03	0.8529	-0.0100	0.05	-0.01	0.2259	0.6485
	8	0.0000	0.0001	0.0000	15.7230	-19.9540	0.0084	0.0000	0.0001	0.0011	0.0697	0.6502	0.0000	0.0000	-0.0002	15.7230	-19.9540	0.0084	0.0000	0.0001	0.0011	0.0697	0.6502
681	5	1.050	2.05	0.0000	-0.0074	-0.02	0.6598	-0.0196	0.05	-0.00	0.1848	0.6278	0.0027	0.0000	0.0000	-0.0074	-0.02	0.6598	-0.0196	0.05	-0.00	0.1848	0.6278
	8	-0.0004	0.0002	0.0000	-2.4402	0.8786	0.0171	0.0004	0.0004	0.0021	0.1185	0.6251	-0.0004	0.0000	-0.0000	-2.4402	0.8786	0.0171	0.0004	0.0004	0.0021	0.1185	0.6251
681	6	1.048	3.11	0.0000	0.0340	-0.02	0.8701	-0.0296	0.06	0.00	0.1727	0.6422	0.0021	0.0000	0.0003	0.0340	-0.02	0.8701	-0.0296	0.06	0.00	0.1727	0.6422
	8	-0.0005	0.0002	0.0000	-2.0880	0.4991	0.0271	0.0007	0.0010	0.0034	0.1348	0.6355	-0.0005	0.0000	-0.0000	-2.0880	0.4991	0.0271	0.0007	0.0010	0.0034	0.1348	0.6355
681	7	1.047	4.14	0.0000	-0.0124	-0.03	0.9322	0.0391	0.06	0.02	0.1606	0.6486	0.0016	0.0000	0.0003	-0.0124	-0.03	0.9322	0.0391	0.06	0.02	0.1606	0.6486
	8	-0.0001	0.0002	0.0000	-9.3752	1.5935	0.0374	0.0010	0.0010	0.0048	0.1370	0.6503	-0.0001	0.0000	-0.0000	-9.3752	1.5935	0.0374	0.0010	0.0010	0.0048	0.1370	0.6503
681	8	1.050	-0.03	0.0000	0.0819	-0.04	1.1169	-0.0020	0.05	-0.03	0.2517	0.5883	0.0016	0.0000	0.0003	0.0819	-0.04	1.1169	-0.0020	0.05	-0.03	0.2517	0.5883
	8	-0.0003	0.0001	0.0000	-2.6552	2.4671	0.0012	-0.0001	-0.0001	0.0001	-0.7707	0.4583	-0.0003	0.0000	-0.0001	-2.6552	2.4671	0.0012	-0.0001	-0.0001	0.0001	-0.7707	0.4583
682	1	1.050	-3.10	0.0000	-0.3509	-0.03	0.7817	-0.0176	0.05	-0.93	0.1700	0.6578	0.0012	0.0000	0.0001	-0.3509	-0.03	0.7817	-0.0176	0.05	-0.93	0.1700	0.6578
	8	-0.0007	0.0001	0.0000	-1.3014	1.3240	-0.0164	-0.0008	-0.0008	-0.0022	0.2642	0.6729	-0.0007	0.0000	-0.0002	-1.3014	1.3240	-0.0164	-0.0008	-0.0008	-0.0022	0.2642	0.6729
682	2	1.048	-0.02	0.0000	0.1066	-0.03	1.1613	0.0076	0.05	0.96	0.2161	0.6606	0.0016	0.0000	0.0003	0.1066	-0.03	1.1613	0.0076	0.05	0.96	0.2161	0.6606
	8	-0.0002	0.0001	0.0000	-4.1307	4.2202	0.0065	0.0000	0.0000	0.0000	0.0181	0.6396	-0.0002	0.0000	-0.0001	-4.1307	4.2202	0.0065	0.0000	0.0000	0.0000	0.0181	0.6396

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																		
682	3	1.048	0.01	-0.00	-0.0151	-0.03	1.02	0.05	0.97	0.2143	0.6556	0.0018	-0.0001	0.0003	0.9188	0.0079	0.0003	0.0010	0.0343	0.6555
	8	0.0000	-0.0000	-0.0002	11.4962	-15.2447	0.0066	0.0000	0.0008											
	9																			
682	4	1.050	1.03	-0.00	-0.0071	-0.03	1.04	0.05	0.97	0.1971	0.6489	0.0024	-0.0002	0.0003	0.7145	0.0163	0.0006	0.0021	0.1089	0.6443
	8	0.0007	-0.0002	-0.0001	-1.5228	0.8054	0.0142	0.0003	0.0018											
	9																			
682	5	1.049	2.08	-0.00	-0.0730	-0.03	1.05	0.05	0.98	0.1726	0.6449	0.0028	-0.0002	0.0003	0.6120	0.0267	0.0009	0.0034	0.1304	0.6385
	8	0.0002	-0.0002	-0.0001	-4.3544	3.0015	0.0237	0.0006	0.0030											
	9																			
682	6	1.050	3.14	-0.00	-0.0394	-0.03	1.07	0.06	1.00	0.1612	0.6544	0.0022	-0.0002	0.0003	0.7465	0.0368	0.0011	0.0048	0.1307	0.6462
	8	0.0009	-0.0002	-0.0000	-1.5447	0.1789	0.0339	0.0008	0.0043											
	9																			
682	7	1.048	4.17	-0.00	-0.2586	-0.03	1.09	0.06	1.02	0.1494	0.6553	0.0023	-0.0001	0.0002	0.5108	0.0466	0.0013	0.0061	0.1331	0.6581
	8	0.0007	-0.0002	-0.0001	-1.5087	-1.0660	0.0447	0.0011	0.0058											
	9																			
682	8	1.047	-0.02	-0.00	-0.0773	-0.03	1.03	0.05	0.97	0.2117	0.6456	0.0020	-0.0002	0.0003	0.8118	0.0076	0.0003	0.0009	0.0150	0.6265
	8	0.0004	-0.0002	-0.0001	-2.2139	1.9032	0.0061	0.0000	0.0007											
	9																			
683	10	0.898	-3.11	-0.00	-0.2700	-0.02	-3.01	0.06	-3.02	0.2145	0.6487	0.0019	-0.0001	0.0003	0.9502	-0.0461	-0.0019	-0.0059	0.2596	0.6344
	8	0.0004	-0.0001	-0.0002	-1.2960	3.6396	-0.0431	-0.0022	-0.0054											
	9																			
683	11	0.897	-1.07	-0.00	-0.1419	-0.02	-2.98	0.05	-2.99	0.2125	0.6687	0.0023	-0.0001	0.0004	1.0280	-0.0244	-0.0010	-0.0032	0.2903	0.6806
	8	0.0009	-0.0001	-0.0002	-0.9279	1.3132	-0.0230	-0.0013	-0.0031											
	9																			
683	12	0.898	-0.04	-0.00	-0.2534	-0.02	-2.96	0.06	-2.97	0.1949	0.7119	0.0036	-0.0001	0.0004	0.6482	-0.0145	-0.0005	-0.0020	0.3329	0.7087
	8	0.0007	-0.0001	-0.0002	-1.0974	1.4824	-0.0138	-0.0009	-0.0019											
	9																			
683	13	0.897	0.47	-0.00	-0.2293	-0.02	-2.96	0.05	-2.97	0.1927	0.7746	0.0040	-0.0001	0.0004	0.6106	-0.0095	-0.0003	-0.0014	0.3623	0.7275
	8	0.0007	-0.0001	-0.0002	-1.1020	1.4539	-0.0097	-0.0007	-0.0014											
	9																			
683	14	0.898	0.96	-0.00	-0.0836	-0.01	-2.94	0.05	-2.97	0.2794	0.8087	0.0035	-0.0002	0.0006	0.8875	-0.0039	-0.0002	-0.0006	0.4100	0.7526
	8	0.0012	-0.0002	-0.0001	-0.8767	0.6423	-0.0060	-0.0004	-0.0009											
	9																			

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd																							
683	15	7	0.898	3.07	-0.004	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	0.2234	0.5360
		6	0.0037	-0.0001	0.0004	-0.2026	0.6481	-0.002	-2.92	0.05	0.0006	0.0002	0.0135	0.0002	0.0017	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.1194	0.6398
		9	-0.0007	0.0002	-0.0001	-1.3176	1.0212	0.0108	0.0108	0.0002	0.0002	0.0002	0.0108	0.0002	0.0013	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002	0.0002
683	16	7	0.897	6.17	-0.000	-0.4361	0.6294	-0.002	-2.86	0.06	0.0018	0.0016	0.0431	0.0016	-2.90	0.06	0.0016	0.0016	0.0016	0.0016	0.0016	0.0016	0.2182	0.6415
		6	0.0003	-0.0002	0.0000	-4.3725	-0.1743	0.0440	0.0440	0.0016	0.0016	0.0016	0.0440	0.0016	0.0055	0.0016	0.0016	0.0016	0.0016	0.0016	0.0016	0.0016	0.1843	0.6292
		9	-0.0003	0.0002	0.0000	-4.3725	-0.1743	0.0440	0.0440	0.0016	0.0016	0.0016	0.0440	0.0016	0.0055	0.0016	0.0016	0.0016	0.0016	0.0016	0.0016	0.0016	0.0016	0.0016
683	17	7	0.898	9.27	-0.000	-3.2094	-4.1118	-0.02	-2.83	0.05	0.0028	0.0026	0.0688	0.0026	-2.85	0.05	0.0028	0.0026	0.0026	0.0026	0.0026	0.0026	0.2077	0.5896
		6	0.0001	-0.0002	0.0001	-0.7594	-0.0923	0.0688	0.0688	0.0026	0.0026	0.0026	0.0688	0.0026	0.0082	0.0026	0.0026	0.0026	0.0026	0.0026	0.0026	0.0026	0.1907	0.5893
		9	-0.0017	0.0002	0.0000	-0.7594	-0.0923	0.0688	0.0688	0.0026	0.0026	0.0026	0.0688	0.0026	0.0082	0.0026	0.0026	0.0026	0.0026	0.0026	0.0026	0.0026	0.0026	0.0026
683	18	7	0.897	12.46	0.000	-0.8208	0.9942	-0.02	-2.81	0.05	0.0030	0.0030	0.0915	0.0030	-2.83	0.05	0.0030	0.0030	0.0030	0.0030	0.0030	0.0030	0.1701	0.5714
		6	0.0011	-0.0001	0.0002	-0.6806	0.3208	0.0915	0.0915	0.0030	0.0030	0.0030	0.0915	0.0030	0.0103	0.0030	0.0030	0.0030	0.0030	0.0030	0.0030	0.0030	0.1658	0.5657
		9	-0.0013	0.0001	0.0000	-0.6806	0.3208	0.0915	0.0915	0.0030	0.0030	0.0030	0.0915	0.0030	0.0103	0.0030	0.0030	0.0030	0.0030	0.0030	0.0030	0.0030	0.0030	0.0030
683	19	7	0.899	-0.06	-0.000	-1.0156	0.7983	-0.02	-2.96	0.06	0.0005	0.0005	0.0136	0.0005	-2.98	0.06	0.0005	0.0005	0.0005	0.0005	0.0005	0.0005	0.1996	0.7192
		6	0.0039	-0.0002	0.0004	-0.2841	0.5846	-0.02	-0.1436	0.0005	0.0005	0.0005	-0.0136	0.0005	0.0021	0.0005	0.0005	0.0005	0.0005	0.0005	0.0005	0.0005	0.3277	0.7167
		9	-0.0011	0.0002	-0.0001	-1.0156	0.7983	-0.02	-0.136	0.0005	0.0005	0.0005	-0.0136	0.0005	0.0019	0.0005	0.0005	0.0005	0.0005	0.0005	0.0005	0.0005	0.0005	0.0005
684	1	7	0.896	-3.10	-0.000	-0.2584	0.8927	-0.02	-1.01	0.06	0.0014	0.0016	0.329	0.0014	-1.04	0.06	0.0014	0.0016	0.0016	0.0016	0.0016	0.0016	0.2217	0.6570
		6	0.0021	-0.0001	0.0003	-0.5779	0.4914	-0.0329	0.0329	0.0014	0.0014	0.0014	0.329	0.0014	0.0043	0.0014	0.0014	0.0014	0.0014	0.0014	0.0014	0.0014	0.2619	0.6519
		9	-0.0021	0.0002	-0.0002	-0.5779	0.4914	-0.0329	0.0329	0.0014	0.0014	0.0014	0.329	0.0014	0.0041	0.0014	0.0014	0.0014	0.0014	0.0014	0.0014	0.0014	0.0014	0.0014
684	2	7	0.899	-1.08	-0.000	-0.1690	1.0221	-0.02	-0.96	0.05	0.0004	0.0004	0.0123	0.0004	-1.02	0.05	0.0004	0.0004	0.0004	0.0004	0.0004	0.0004	0.1798	0.7165
		6	0.0024	-0.0000	0.0005	-0.9257	1.1886	-0.0114	-0.0114	0.0004	0.0004	0.0004	-0.0114	0.0004	0.0015	0.0004	0.0004	0.0004	0.0004	0.0004	0.0004	0.0004	0.3159	0.6952
		9	-0.0010	0.0001	-0.0002	-0.9257	1.1886	-0.0114	-0.0114	0.0004	0.0004	0.0004	-0.0114	0.0004	0.0015	0.0004	0.0004	0.0004	0.0004	0.0004	0.0004	0.0004	0.0004	0.0004
684	3	7	0.897	-0.05	-0.000	-0.0831	1.0423	-0.02	-0.95	0.05	0.0001	0.0001	0.0020	0.0001	-1.01	0.05	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.3071	1.0307
		6	0.0028	-0.0000	0.0005	-0.9372	1.1870	-0.0039	-0.0039	0.0001	0.0001	0.0001	-0.0039	0.0001	0.0004	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.4474	0.7940
		9	-0.0009	0.0001	-0.0002	-0.9372	1.1870	-0.0039	-0.0039	0.0001	0.0001	0.0001	-0.0039	0.0001	0.0006	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001
684	4	7	0.899	0.50	-0.000	-0.1690	1.0002	-0.02	-0.94	0.05	0.0001	0.0001	0.0017	0.0001	-0.99	0.05	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.4668	0.4158
		6	0.0038	-0.0001	0.0005	-0.9285	1.0002	-0.0017	-0.0017	0.0001	0.0001	0.0001	-0.0017	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	1.2442	1.4053
		9	-0.0010	0.0001	-0.0002	-0.9285	1.0002	-0.0017	-0.0017	0.0001	0.0001	0.0001	-0.0017	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001	0.0001
684	5	7	0.896	0.99	-0.000	-0.1114	0.8345	-0.02	-0.93	0.05	0.0003	0.0003	0.0055	0.0003	-0.99	0.05	0.0003	0.0003	0.0003	0.0003	0.0003	0.0003	0.3460	0.5911
		6	0.0035	-0.0000	0.0006	-0.8763	0.9116	-0.0039	-0.0039	0.0003	0.0003	0.0003	-0.0039	0.0003	0.0006	0.0003	0.0003	0.0003	0.0003	0.0003	0.0003	0.0003	0.0431	0.6793
		9	-0.0010	0.0001	-0.0001	-0.8763	0.9116	-0.0039	-0.0039	0.0003	0.0003	0.0003	-0.0039	0.0003	0.0006	0.0003	0.0003	0.0003	0.0003	0.0003	0.0003	0.0003	0.0003	0.0003
684	6	7	0.901	3.09	-0.000	-0.1855	0.7893	-0.02	-0.90	0.05	0.0011	0.0011	0.0260	0.0011	-0.97	0.05	0.0011	0.0011	0.0011	0.0011	0.0011	0.0011	0.2269	0.6473
		6	0.0031	-0.0001	0.0004	-1.4139	1.5611	-0.0260	-0.0260	0.0011	0.0011	0.0011	-0.0260	0.0011	0.0033	0.0011	0.0011	0.0011	0.0011	0.0011	0.0011	0.0011	0.1734	0.6554
		9	-0.0006	0.0001	-0.0001	-1.4139	1.5611	-0.0260	-0.0260	0.0011	0.0011	0.0011	-0.0260	0.0011	0.0030	0.0011	0.0011	0.0011	0.0011	0.0011	0.0011	0.0011	0.0011	0.0011

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd																			
684	7	0.898	6.18	-0.000	-1.6219	-0.02	-0.86	0.06	-0.92	0.2143	0.6143									
	8	-0.0001	0.0000	-0.0000	-3.6676	0.5402	0.0544	0.0023	0.0066	0.1915	0.6238									
	9	-0.00004	0.0003	0.0000		-0.5549	0.0559	0.0021	0.0069											
684	8	0.898	9.28	-0.000	1.0099	-0.02	-0.83	0.05	-0.88	0.1974	0.5793									
	8	-0.0000	-0.0000	0.0000	-1.2692	3.1359	0.767	0.0030	0.0088	0.1849	0.5780									
	9	-0.00009	0.0002	-0.0000		0.0938	0.0789	0.0029	0.0091											
684	9	0.900	12.46	0.000	-1.1446	-0.02	-0.81	0.05	-0.86	0.1622	0.5664									
	8	0.0011	-0.0002	0.0001	-0.5056	0.5351	0.0956	0.0031	0.0108	0.1601	0.5576									
	9	-0.0016	0.0001	-0.0001		0.4673	0.0973	0.0031	0.0108											
684	10	0.900	-0.04	-0.000	-0.1890	-0.02	-0.95	0.05	-1.00	0.2105	0.9094									
	8	0.0030	-0.0001	0.0005	-0.9844	0.8764	-0.023	-0.0000	-0.0004	0.4477	0.8336									
	9	-0.0010	0.0001	-0.0002		1.0950	-0.0038	-0.0003	-0.0006											
685	1	0.900	-3.10	-0.000	-0.3425	-0.02	-0.11	0.05	-0.07	0.2111	0.6538									
	8	0.0019	-0.0001	0.0003	-0.5568	0.8566	-0.273	-0.0013	-0.0035	0.2727	0.6670									
	9	-0.0016	0.0001	-0.0003		0.9272	-0.0251	-0.0013	-0.0035											
685	2	0.898	-1.07	-0.000	-0.2425	-0.02	-0.08	0.05	-0.04	0.2084	0.7397									
	8	0.0028	-0.0001	0.0004	-0.9279	0.8501	-0.060	-0.0002	-0.0008	0.5860	0.7370									
	9	-0.0009	0.0001	-0.0002		1.3132	-0.0061	-0.0004	-0.0009											
685	3	0.898	-0.03	-0.000	-0.2065	-0.02	-0.06	0.06	-0.02	0.4749	0.4819									
	8	0.0033	-0.0001	0.0005	-1.1523	0.7998	0.020	0.0001	0.0001	-0.3265	0.4757									
	9	-0.0006	0.0001	-0.0002		2.1067	0.0011	-0.0000	0.0001											
685	4	0.898	0.52	-0.000	-0.0346	-0.01	-0.06	0.05	-0.02	0.3452	0.6003									
	8	0.0029	-0.0000	0.0006	-1.0571	1.0336	0.061	0.0001	0.0007	0.0907	0.6584									
	9	-0.0009	0.0001	-0.0002		1.0905	0.0056	0.0001	0.0007											
685	5	0.900	1.00	-0.000	-0.1242	-0.02	-0.05	0.05	-0.01	0.2445	0.6459									
	8	0.0035	-0.0000	0.0005	-1.0034	0.8222	0.115	0.0002	0.0014	0.1541	0.6620									
	9	-0.0008	0.0001	-0.0002		1.2746	0.0095	0.0002	0.0012											
685	6	0.897	3.09	-0.000	-0.0746	-0.02	-0.01	0.05	-0.01	0.2201	0.6445									
	8	0.0026	-0.0000	0.0005	-1.0859	0.9643	0.0322	0.0011	0.0038	0.1674	0.6457									
	9	-0.0010	0.0002	-0.0001		0.7190	0.0300	0.0011	0.0038											
685	7	0.897	6.18	-0.000	-0.1731	-0.02	-0.03	0.06	-0.05	0.2167	0.6039									
	8	0.0004	-0.0000	0.0000	-1.6239	0.4912	0.0590	0.0025	0.0073	0.1935	0.6059									
	9	-0.0010	0.0003	-0.0000		-0.3675	0.0603	0.0023	0.0073											

← See Key →

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key													
685	8	7	0.898	9.30	-0.000	-1.4059	-0.02	0.05	0.05	0.08	0.1898	0.5742			
		8	0.0001	-0.0000	0.0000	2.4407	0.0607	0.0030	0.0030	0.0092	0.1833	0.5698			
		9	-0.00014	0.0002	-0.0000	-0.6839	0.0825	0.0030	0.0030	0.0094	0.1524	0.5533			
685	9	7	0.896	12.45	-0.001	-1.0855	-0.02	0.05	0.05	0.08	0.1539	0.5622			
		8	0.0012	-0.0002	0.0001	0.5354	0.0978	0.0030	0.0030	0.0109	0.1524	0.5533			
		9	-0.0026	0.0002	0.0000	-0.5178	0.0998	0.0030	0.0030	0.0110	0.1524	0.5533			
685	10	7	0.897	-0.04	-0.00	-0.02	-0.06	0.05	0.05	-0.02	0.3186	0.3411			
		8	0.0037	-0.0002	0.0004	0.6020	0.0025	0.0001	0.0001	0.0001	0.2160	0.3908			
		9	-0.0016	0.0002	-0.0001	0.4852	0.0011	-0.0000	-0.0000	0.0000	-0.2160	0.3908			
686	1	7	0.897	-3.06	-0.00	-0.02	0.97	0.05	0.05	0.92	0.1948	0.6746			
		8	0.0020	-0.0001	0.0003	0.8103	-0.0203	-0.0007	-0.0010	0.0027	0.2797	0.6703			
		9	-0.0016	0.0002	-0.0002	0.6011	-0.0185	-0.0010	-0.0010	-0.0024	0.2797	0.6703			
686	2	7	0.899	-1.04	-0.00	-0.02	1.00	0.05	0.05	0.95	-1.3916	1.8738			
		8	0.0025	-0.0000	0.0005	0.1820	-0.0002	0.0000	0.0000	0.0001	-0.4998	0.8168			
		9	-0.0011	0.0001	-0.0002	0.8682	-0.0014	-0.0001	-0.0001	-0.0002	-0.4998	0.8168			
686	3	7	0.896	-0.01	-0.00	-0.02	1.01	0.05	0.05	0.95	0.3575	0.6373			
		8	0.0029	-0.0000	0.0005	0.9605	0.0072	0.0005	0.0005	0.0009	0.1340	0.6661			
		9	-0.0006	0.0001	-0.0002	1.3221	0.0066	0.0001	0.0001	0.0008	0.1340	0.6661			
686	4	7	0.898	0.52	-0.00	-0.02	1.02	0.05	0.05	0.97	0.2382	0.6715			
		8	0.0034	-0.0000	0.0005	0.8233	0.0135	0.0006	0.0006	0.0018	0.1824	0.6675			
		9	-0.0008	0.0001	-0.0002	1.3986	0.0108	0.0003	0.0003	0.0014	0.1824	0.6675			
686	5	7	0.896	1.03	-0.00	-0.02	1.03	0.05	0.05	0.97	0.2343	0.6504			
		8	0.0040	-0.0001	0.0005	0.6466	0.0186	0.0008	0.0008	0.0024	0.2343	0.6504			
		9	-0.0009	0.0001	-0.0002	1.1076	0.0164	0.0005	0.0005	0.0021	0.1604	0.6585			
686	6	7	0.898	3.09	-0.00	-0.02	1.07	0.05	0.05	1.00	0.2199	0.6454			
		8	0.0031	-0.0001	0.0004	0.6627	0.0397	0.0017	0.0017	0.0051	0.1944	0.6509			
		9	-0.0005	0.0002	-0.0001	1.5681	0.0369	0.0014	0.0014	0.0048	0.1944	0.6509			
686	7	7	0.899	6.20	-0.00	-0.02	1.11	0.06	0.06	1.04	0.2120	0.5970			
		8	0.0003	0.0000	0.0000	0.5161	0.0639	0.0027	0.0027	0.0076	0.2120	0.5970			
		9	-0.0006	0.0002	-0.0000	-0.4917	0.0648	0.0025	0.0025	0.0077	0.1964	0.5965			
686	8	7	0.899	9.30	-0.00	-0.02	1.13	0.05	0.05	1.06	0.1817	0.5754			
		8	0.0010	-0.0001	0.0001	0.8337	0.0850	0.0030	0.0030	0.0097	0.1793	0.5650			
		9	-0.0012	0.0002	0.0000	-0.9773	0.0858	0.0030	0.0030	0.0096	0.1793	0.5650			

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																													
686	9	7	0.899	12.46	0.00	-1.0624	-0.02	1.14	0.05	1.07	0.1423	0.5660	8	0.0011	-0.0002	0.0001	-0.5412	0.6781	0.1003	0.0028	0.0113	0.1422	0.5560	9	-0.0019	0.0000	-0.0000	0.2126	0.1026	0.0029	0.0114
686	10	7	0.900	-0.0001	-0.0002	-0.2217	-0.02	1.01	0.05	0.95	0.3522	0.6173	6	0.0032	-0.0001	0.0005	1.1447	0.0073	0.0005	0.0009	0.0009	0.1579	0.6173	9	-0.0009	-0.0001	-0.0002	0.8153	0.0067	0.0001	0.0009
686	11	7	0.898	-3.08	-0.00	-0.3738	-0.02	2.96	0.05	3.03	0.1516	0.6759	6	0.0018	-0.0001	0.0003	0.9115	0.0076	0.0002	-0.0010	0.0009	0.3214	0.6759	9	-0.0017	-0.0001	-0.0003	0.9273	-0.0065	-0.0004	-0.0009
686	12	7	0.897	-1.03	-0.00	-0.2341	-0.02	2.99	0.05	3.06	0.2735	0.6997	6	0.0029	-0.0001	0.0004	0.8567	0.0098	0.0005	0.0015	0.0013	0.2032	0.6997	9	-0.0013	-0.0002	-0.0001	0.7211	0.0098	0.0004	0.0013
687	1	7	0.899	0.00	-0.00	-0.2664	-0.02	3.01	0.05	3.08	0.2392	0.6562	6	0.0034	-0.0001	0.0005	0.7298	0.0207	0.0009	0.0027	0.0027	0.1778	0.6562	9	-0.0007	-0.0002	-0.0001	1.5372	0.0203	0.0007	0.0027
687	2	7	0.899	0.52	-0.00	-0.1528	-0.02	3.01	0.05	3.09	0.2378	0.6601	6	0.0034	-0.0001	0.0005	0.7905	0.0248	0.0012	0.0034	0.0032	0.1913	0.6601	9	-0.0006	-0.0002	-0.0001	1.7905	0.0248	0.0009	0.0032
687	3	7	0.899	1.04	-0.00	-0.1218	-0.02	3.02	0.05	3.10	0.2365	0.6526	6	0.0032	-0.0001	0.0005	0.806	0.0289	0.0014	0.0041	0.0039	0.1936	0.6526	9	-0.0005	-0.0002	-0.0001	2.5403	0.0289	0.0011	0.0039
687	4	7	0.897	3.11	-0.00	-0.0714	-0.02	3.05	0.05	3.13	0.2202	0.6216	6	0.0021	-0.0000	0.0003	0.8333	0.0512	0.0022	0.0063	0.0063	0.1917	0.6216	9	-0.0012	-0.0001	-0.0001	0.4360	0.0507	0.0019	0.0063
687	5	7	0.899	6.19	-0.00	-2.3798	-0.02	3.09	0.06	3.15	0.2060	0.5843	6	0.0001	0.0000	0.0000	0.6320	0.0722	0.0029	0.0084	0.0086	0.1978	0.5843	9	-0.0004	-0.0002	-0.0000	0.4750	0.0738	0.0029	0.0086
687	6	7	0.901	9.31	-0.00	-0.7285	-0.02	3.11	0.05	3.17	0.1729	0.5660	6	0.0015	-0.0002	0.0001	0.488	0.0907	0.0031	0.0102	0.0103	0.1687	0.5660	9	-0.0025	-0.0003	-0.0001	0.2456	0.0932	0.0031	0.0103
687	7	7	0.899	12.46	0.00	-3.2970	-0.02	3.11	0.05	3.17	0.1265	0.5591	6	0.0002	-0.0001	0.0002	3.8610	0.1040	0.0026	0.0118	0.0120	0.1279	0.5591	9	-0.0003	-0.0000	-0.0002	4.5697	0.1073	0.0027	0.0120

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																
687	8	7	0.899	-0.001	-0.00	-0.02	3.01	0.05	3.08	0.2431	0.6593	0.0036	0.0004	0.6855	0.0205	0.0009	0.0027	0.1740	0.6615
		8	-0.0011	0.0002	-0.0002	0.8746	0.0199	0.0006	0.0026										
		9																	
68A	1	7	0.89A	-3.007	-0.00	-0.01	6.01	0.05	6.00	0.3279	0.7180	0.0014	0.0004	1.4097	0.0088	0.0005	0.0012	0.2421	0.7123
		8	-0.000A	0.0001	-0.0004	2.3169	0.0084	0.0004	0.0012										
		9																	
68A	2	7	0.899	-1.00	-0.00	-0.02	6.04	0.05	6.02	0.2401	0.6668	0.0026	0.0005	0.9671	0.0300	0.0014	0.0039	0.2010	0.6668
		8	-0.0006	0.0001	-0.0002	2.1061	0.0298	0.0012	0.0039										
		9																	
688	3	7	0.899	-0.001	-0.00	-0.02	6.06	0.05	6.03	0.2265	0.6486	0.0028	0.0005	1.0551	0.0407	0.0018	0.0052	0.2207	0.6547
		8	-0.0007	0.0001	-0.0002	1.3945	0.0396	0.0015	0.0051										
		9																	
68A	4	7	0.899	0.56	-0.00	-0.02	6.06	0.05	6.05	0.2283	0.6448	0.0038	0.0005	0.6728	0.0450	0.0020	0.0058	0.2203	0.6448
		8	-0.0004	0.0001	-0.0002	2.8204	0.0440	0.0017	0.0056										
		9																	
68A	5	7	0.899	1.02	-0.00	-0.02	6.07	0.05	6.05	0.2251	0.6284	0.0032	0.0005	0.9043	0.0496	0.0022	0.0062	0.1966	0.6399
		8	-0.0005	0.0001	-0.0002	1.9043	0.0489	0.0019	0.0062										
		9																	
688	6	7	0.899	3.12	-0.00	-0.02	6.09	0.05	6.08	0.2149	0.5972	0.0019	0.0002	0.7434	0.0647	0.0027	0.0077	0.2026	0.5972
		8	-0.0009	0.0002	-0.0001	0.6407	0.0644	0.0026	0.0077										
		9																	
68A	7	7	0.899	6.24	-0.00	-0.02	6.12	0.05	6.11	0.1842	0.5730	0.0006	0.0001	0.8628	0.0843	0.0031	0.0095	0.1850	0.5663
		8	-0.000A	0.0002	-0.0001	-0.6345	0.0845	0.0031	0.0095										
		9																	
68A	8	7	0.899	9.30	-0.00	-0.02	6.13	0.05	6.10	0.1459	0.5645	0.0013	0.0001	0.8910	0.0978	0.0028	0.0110	0.1455	0.5577
		8	-0.0006	0.0001	-0.0001	-1.3101	0.1000	0.0029	0.0110										
		9																	
688	9	7	0.898	12.46	0.00	-0.02	6.12	0.05	6.10	0.1129	0.5676	0.0013	0.0001	0.4043	0.1076	0.0024	0.0122	0.1178	0.5530
		8	-0.0010	0.0001	-0.0001	0.6521	0.1102	0.0025	0.0122										
		9																	
68A	10	7	0.899	-0.00	-0.00	-0.02	6.05	0.05	6.04	0.2230	0.6456	0.0031	0.0001	0.8259	0.0404	0.0018	0.0051	0.2230	0.6456
		8	-0.0009	0.0001	-0.0002	1.0905	0.0396	0.0015	0.0051										
		9																	

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key										
689	1	7	0.897	-3.02	-0.00	-0.3902	-0.02	9.07	0.05	9.02	0.2479	0.6737
		8	0.0023	-0.0001	0.0003	-0.5763	0.7642	0.0280	0.0013	0.0037	0.2021	0.6750
		9	-0.0015	0.0001	-0.0002	-0.5763	0.8802	0.0287	0.0011	0.0038		
689	2	7	0.899	-0.99	-0.00	-0.3293	-0.01	9.11	0.05	9.05	0.2283	0.6333
		8	0.0028	-0.0001	0.0004	-1.3989	0.8309	0.0478	0.0021	0.0060	0.2024	0.6463
		9	-0.0005	0.0001	-0.0002	-1.3989	2.5792	0.0481	0.0019	0.0062		
689	3	7	0.897	0.00	-0.00	-0.2528	-0.02	9.12	0.05	9.06	0.2233	0.6142
		8	0.0029	-0.0001	0.0005	-0.7848	0.8663	0.0561	0.0025	0.0068	0.1998	0.6227
		9	-0.0012	0.0001	-0.0001	-0.7848	0.7396	0.0559	0.0022	0.0069		
689	4	7	0.900	0.58	-0.00	0.0273	-0.02	9.12	0.05	9.06	0.2203	0.6058
		8	0.0019	0.0000	0.0005	-2.8009	1.4578	0.0597	0.0026	0.0072	0.2015	0.6140
		9	-0.0002	0.0001	-0.0002	-2.8009	7.3258	0.0593	0.0023	0.0072		
689	5	7	0.901	1.07	-0.00	-0.0416	-0.02	9.13	0.05	9.07	0.2192	0.5983
		8	0.0022	-0.0000	0.0005	-1.1275	1.580	0.0632	0.0027	0.0075	0.2033	0.6032
		9	-0.0006	0.0001	-0.0002	-1.1275	1.8948	0.0627	0.0025	0.0075		
689	6	7	0.899	3.14	-0.00	-0.0359	-0.01	9.14	0.05	9.09	0.2064	0.5728
		8	0.0018	-0.0000	0.0002	-1.0238	0.5575	0.0763	0.0031	0.0087	0.1976	0.5728
		9	-0.0011	0.0002	-0.0001	-1.0238	0.5576	0.0763	0.0030	0.0087		
689	7	7	0.897	6.24	-0.00	-0.4566	-0.01	9.15	0.06	9.10	0.1648	0.5562
		8	0.0011	-0.0001	0.0002	-2.0975	0.8811	0.0922	0.0030	0.0103	0.1644	0.5562
		9	-0.0007	0.0003	-0.0000	-2.0975	-0.3061	0.0943	0.0031	0.0105		
689	8	7	0.897	9.33	-0.00	-0.8619	-0.02	9.16	0.05	9.10	0.1237	0.5571
		8	0.0010	-0.0001	0.0001	-5.0397	0.9656	0.1020	0.0025	0.0115	0.1243	0.5571
		9	-0.0001	0.0001	-0.0001	-5.0397	4.0567	0.1044	0.0025	0.0116		
689	9	7	0.899	12.48	0.00	-4.7608	-0.02	9.17	0.05	9.11	0.1109	0.5501
		8	0.0001	-0.0001	0.0001	-4.7608	3.7320	0.1127	0.0025	0.0126	0.1152	0.5501
		9	-0.0017	0.0001	-0.0001	-4.7608	0.5161	0.1155	0.0026	0.0127		
689	10	7	0.897	0.02	-0.00	-0.3077	-0.02	9.11	0.05	9.06	0.2319	0.6102
		8	0.0034	-0.0002	0.0004	-0.9202	0.6674	0.0562	0.0024	0.0068	0.2319	0.6102
		9	-0.0011	0.0002	-0.0002	-0.9202	0.8746	0.0559	0.0022	0.0069		
690	3	7	0.899	-3.00	-0.00	-0.4361	-0.02	12.05	0.05	12.06	0.2458	0.6522
		8	0.0017	-0.0001	0.0004	-0.4361	1.1497	0.0450	0.0022	0.0058	0.2458	0.6522
		9	-0.0008	0.0000	-0.0003	-0.4361	2.1315	0.0467	0.0019	0.0061		

TABLE VI
 CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																	
690	4	7 8 9	0.900 0.0030 -0.0009	-0.0002 0.0001 0.0001	-0.0002 0.0001 0.0001	-0.0003 0.0003 0.0003	-0.4578 -0.6569 -0.4968	-0.02 0.6576 1.1425	12.08 0.0611 0.0624	0.05 0.0028 0.0025	12.08 0.0075 0.0076	0.2298 0.2061 0.6132	0.2298 0.2061 0.6132	0.2298 0.2061 0.6132	0.2298 0.2061 0.6132	0.2298 0.2061 0.6132	0.2298 0.2061 0.6132	0.2298 0.2061 0.6132	0.2298 0.2061 0.6132	0.2298 0.2061 0.6132
690	5	7 8 9	0.899 0.0034 -0.0017	-0.0002 0.0001 0.0001	-0.0002 0.0001 0.0001	-0.0003 0.0003 0.0003	-0.3570 -0.4968 -0.4968	-0.02 0.5298 0.4289	12.09 0.0681 0.0687	0.05 0.0030 0.0028	12.09 0.0081 0.0081	0.2219 0.2042 0.5977	0.2219 0.2042 0.5977	0.2219 0.2042 0.5977	0.2219 0.2042 0.5977	0.2219 0.2042 0.5977	0.2219 0.2042 0.5977	0.2219 0.2042 0.5977	0.2219 0.2042 0.5977	0.2219 0.2042 0.5977
690	6	7 8 9	0.898 0.0022 -0.0004	-0.0000 0.0000 0.0000	-0.0002 0.0004 -0.0002	-0.0004 0.0004 -0.0002	-0.1860 -1.0083 -1.0083	-0.02 0.9153 2.8564	12.09 0.0714 0.0721	0.05 0.0031 0.0029	12.10 0.0084 0.0084	0.2231 0.2015 0.5879	0.2231 0.2015 0.5879	0.2231 0.2015 0.5879	0.2231 0.2015 0.5879	0.2231 0.2015 0.5879	0.2231 0.2015 0.5879	0.2231 0.2015 0.5879	0.2231 0.2015 0.5879	0.2231 0.2015 0.5879
690	7	7 8 9	0.899 0.0024 -0.0012	-0.0000 0.0000 0.0001	-0.0000 0.0004 -0.0001	-0.0004 0.0004 -0.0001	-0.0887 -0.6618 -0.6618	-0.02 0.9220 0.6171	12.10 0.0745 0.0750	0.05 0.0032 0.0030	12.10 0.0086 0.0087	0.2165 0.2000 0.5819	0.2165 0.2000 0.5819	0.2165 0.2000 0.5819	0.2165 0.2000 0.5819	0.2165 0.2000 0.5819	0.2165 0.2000 0.5819	0.2165 0.2000 0.5819	0.2165 0.2000 0.5819	0.2165 0.2000 0.5819
690	8	7 8 9	0.899 0.0017 -0.0014	-0.0002 0.0000 0.0002	-0.0000 0.0002 -0.0000	-0.0002 0.0002 -0.0000	0.0467 0.7615 -0.7615	-0.02 0.7623 0.3540	12.11 0.0864 0.0870	0.05 0.0031 0.0030	12.11 0.0099 0.0098	0.1822 0.1778 0.5738	0.1822 0.1778 0.5738	0.1822 0.1778 0.5738	0.1822 0.1778 0.5738	0.1822 0.1778 0.5738	0.1822 0.1778 0.5738	0.1822 0.1778 0.5738	0.1822 0.1778 0.5738	0.1822 0.1778 0.5738
690	9	7 8 9	0.898 0.0013 -0.0001	-0.0002 0.0001 0.0002	-0.0000 0.0002 -0.0000	-0.0002 0.0002 -0.0000	-0.5764 -6.1183 -6.1183	-0.02 0.8675 1.3710	12.11 0.0969 0.0992	0.05 0.0028 0.0028	12.11 0.0110 0.0111	0.1455 0.1419 0.5687	0.1455 0.1419 0.5687	0.1455 0.1419 0.5687	0.1455 0.1419 0.5687	0.1455 0.1419 0.5687	0.1455 0.1419 0.5687	0.1455 0.1419 0.5687	0.1455 0.1419 0.5687	0.1455 0.1419 0.5687
690	10	7 8 9	0.898 0.0009 -0.0009	-0.0001 0.0001 0.0002	-0.0002 0.0002 -0.0000	-0.0002 0.0002 -0.0000	-0.8702 -1.0675 -1.0675	-0.02 1.2966 0.2316	12.11 0.1056 0.1079	0.05 0.0024 0.0025	12.11 0.0120 0.0120	0.1180 0.1179 0.5696	0.1180 0.1179 0.5696	0.1180 0.1179 0.5696	0.1180 0.1179 0.5696	0.1180 0.1179 0.5696	0.1180 0.1179 0.5696	0.1180 0.1179 0.5696	0.1180 0.1179 0.5696	0.1180 0.1179 0.5696
690	11	7 8 9	0.898 0.0008 -0.0011	-0.0001 0.0001 0.0001	-0.0002 0.0002 -0.0000	-0.0002 0.0002 -0.0000	-0.8715 -0.7016 -0.7016	-0.02 1.3488 0.4027	12.12 0.1161 0.1192	0.05 0.0025 0.0023	12.12 0.0132 0.0134	0.1090 0.0986 0.5685	0.1090 0.0986 0.5685	0.1090 0.0986 0.5685	0.1090 0.0986 0.5685	0.1090 0.0986 0.5685	0.1090 0.0986 0.5685	0.1090 0.0986 0.5685	0.1090 0.0986 0.5685	0.1090 0.0986 0.5685
690	12	7 8 9	0.896 0.0025 -0.0013	-0.0000 0.0002 0.0002	-0.0000 0.0005 -0.0001	-0.0000 0.0005 -0.0001	-0.1600 -0.7536 -0.7536	-0.02 0.9993 0.5497	12.09 0.0685 0.0690	0.05 0.0030 0.0027	12.10 0.0081 0.0081	0.2194 0.2019 0.5922	0.2194 0.2019 0.5922	0.2194 0.2019 0.5922	0.2194 0.2019 0.5922	0.2194 0.2019 0.5922	0.2194 0.2019 0.5922	0.2194 0.2019 0.5922	0.2194 0.2019 0.5922	0.2194 0.2019 0.5922
691	1	7 8 9	0.898 0.0025 -0.0009	-0.0002 0.0001 0.0001	-0.0003 0.0003 -0.0002	-0.0003 0.0003 -0.0002	-0.4660 -0.7509 -0.7509	-0.02 0.7573 1.5952	15.10 0.0601 0.0618	0.05 0.0027 0.0025	15.08 0.0074 0.0076	0.2290 0.2057 0.6154	0.2290 0.2057 0.6154	0.2290 0.2057 0.6154	0.2290 0.2057 0.6154	0.2290 0.2057 0.6154	0.2290 0.2057 0.6154	0.2290 0.2057 0.6154	0.2290 0.2057 0.6154	0.2290 0.2057 0.6154
691	2	7 8 9	0.897 0.0025 -0.0010	-0.0001 0.0001 0.0001	-0.0004 0.0004 -0.0001	-0.0004 0.0004 -0.0001	-0.3366 -0.7728 -0.7728	-0.02 0.8315 0.9438	15.12 0.0732 0.0745	0.05 0.0032 0.0030	15.09 0.0085 0.0087	0.2209 0.2018 0.5859	0.2209 0.2018 0.5859	0.2209 0.2018 0.5859	0.2209 0.2018 0.5859	0.2209 0.2018 0.5859	0.2209 0.2018 0.5859	0.2209 0.2018 0.5859	0.2209 0.2018 0.5859	0.2209 0.2018 0.5859

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																	
691	3	0.898	0.04	-0.003	-0.02	15.13	0.06	15.10	0.2031	0.5700	0.0029	-0.0001	0.0003	0.7761	0.0793	0.0032	0.0091	0.1889	0.5718
	8	-0.0011	-0.0001	0.0001	-0.2774	0.0803	0.0030	0.0091											
	9				-0.7813	0.0803	0.0030	0.0091											
691	4	0.899	0.56	-0.003	-0.02	15.13	0.05	15.11	0.1927	0.5749	0.0025	-0.0001	0.0003	0.6895	0.0825	0.0031	0.0094	0.1853	0.5681
	8	-0.0009	-0.0001	0.0001	-0.1999	0.0826	0.0030	0.0094											
	9				-0.8797	0.0826	0.0030	0.0094											
691	5	0.899	1.05	-0.003	-0.02	15.13	0.05	15.11	0.1832	0.5751	0.0022	-0.0001	0.0003	0.8014	0.0851	0.0030	0.0096	0.1795	0.5636
	8	-0.0009	-0.0001	0.0001	-0.0797	0.0851	0.0030	0.0096											
	9				-0.9382	0.0851	0.0030	0.0096											
691	6	0.899	3.16	-0.002	-0.02	15.13	0.05	15.10	0.1638	0.5677	0.0014	-0.0003	0.0002	0.8864	0.0935	0.0030	0.0106	0.1578	0.5559
	8	0.0014	0.0000	0.0003	0.0322	0.0935	0.0030	0.0106											
	9	-0.0007	0.0000	-0.0003	0.5718	0.0935	0.0030	0.0106											
691	7	0.897	6.23	-0.002	-0.02	15.13	0.06	15.10	0.1248	0.5705	0.0011	-0.0002	0.0000	1.1841	0.1016	0.0025	0.0116	0.1217	0.5583
	8	0.0011	-0.0002	0.0000	-0.4859	0.1016	0.0025	0.0116											
	9	-0.0003	0.0002	0.0000	-3.7933	0.1016	0.0025	0.0116											
691	8	0.899	9.32	-0.001	-0.02	15.14	0.05	15.11	0.1161	0.5673	0.0013	-0.0002	0.0000	0.6831	0.1134	0.0026	0.0125	0.1175	0.5528
	8	0.0013	-0.0002	0.0000	-0.7475	0.1134	0.0026	0.0125											
	9	-0.0014	0.0002	0.0000	-0.9270	0.1134	0.0026	0.0125											
691	9	0.898	12.49	0.001	-0.02	15.14	0.06	15.11	0.0970	0.5701	0.0010	-0.0001	0.0000	0.9871	0.1197	0.0022	0.0136	0.0916	0.5607
	8	0.0010	-0.0001	0.0001	-0.8149	0.1197	0.0022	0.0136											
	9	-0.0017	0.0001	-0.0001	-0.5226	0.1197	0.0022	0.0136											
691	10	0.896	0.04	-0.003	-0.02	15.12	0.06	15.10	0.2011	0.5755	0.0020	-0.0003	0.0003	0.9587	0.1214	0.0030	0.0092	0.1887	0.5707
	8	0.0020	-0.0002	0.0001	-0.1629	0.0806	0.0030	0.0092											
	9	-0.0013	0.0002	-0.0001	-0.7535	0.0806	0.0030	0.0092											
692	1	0.899	3.07	-0.0015	-0.02	2.96	2.94	3.01	0.1130	0.6750	0.0121	-0.0010	0.0015	0.6268	0.0104	0.0002	-0.0051	0.2686	0.6419
	8	-0.0121	-0.0009	0.0012	0.3742	-0.0404	-0.0002	-0.0051											
	9	0.0108	0.0010	-0.0012	0.4890	-0.0404	-0.0002	-0.0051											
692	2	0.898	-0.03	-0.0013	-0.02	3.01	2.94	2.97	0.2842	0.6953	0.0110	-0.0008	0.0013	0.6278	0.0175	0.0009	-0.0024	0.3593	0.7356
	8	-0.0110	-0.0008	-0.0013	0.4010	0.0175	0.0009	-0.0024											
	9	0.0111	0.0010	-0.0014	0.4767	-0.0119	-0.0008	-0.0012											
692	3	0.899	0.51	-0.0013	-0.02	3.01	2.94	2.97	0.2570	0.6663	0.0105	-0.0009	0.0013	0.6440	0.0231	0.0011	-0.0030	0.4097	0.7583
	8	-0.0105	-0.0009	-0.0013	0.4281	0.0231	0.0011	-0.0030											
	9	-0.0111	0.0010	-0.0014	0.4788	-0.0079	-0.0006	-0.0012											

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																			
692	4	0.899	1.02	-0.00	0.3984	-3.01	3.03	2.94	-2.96	0.2459	0.6580	-0.0107	-0.0008	-0.0013	0.3984	0.6246	0.0287	0.0014	0.0037	0.4601	0.7878
	8	-0.0119	0.0010	0.0013	0.4186	0.5763	-0.0045	-0.0004	-0.0007												
	9																				
692	5	0.899	2.06	-0.00	0.4244	-3.01	3.04	2.94	-2.95	0.2366	0.6510	-0.0107	-0.0009	-0.0014	0.4244	0.6576	0.0390	0.0018	0.0050	0.2366	0.5288
	8	-0.0111	0.0010	0.0014	0.4738	0.6606	0.0036	-0.0000	0.0003	-0.0555											
	9																				
692	6	0.899	3.09	-0.00	0.3734	-3.02	3.06	2.94	-2.94	0.2328	0.6261	-0.0119	-0.0008	-0.0015	0.3734	0.6518	0.0481	0.0022	0.0060	0.2328	0.6039
	8	-0.0122	0.0010	0.0014	0.4102	0.5739	0.0138	0.0003	0.0016	0.1099											
	9																				
692	7	0.899	4.10	-0.00	0.2632	-3.01	3.07	2.94	-2.92	0.2293	0.6088	-0.0141	-0.0007	-0.0016	0.2632	0.5994	0.0557	0.0025	0.0067	0.2293	0.6189
	8	-0.0110	0.0011	0.0017	0.5371	0.7762	0.0249	0.0006	0.0030	0.1334											
	9																				
692	8	0.900	-0.003	-0.00	0.3732	-3.01	3.01	2.94	-2.97	0.2661	0.6736	-0.0113	-0.0008	-0.0013	0.3732	0.5981	0.0179	0.0009	0.0024	0.2661	0.7313
	8	-0.0118	0.0010	0.0013	0.4431	0.5864	-0.0119	-0.0008	-0.0017	0.3524											
	9																				
693	1	0.896	-3.10	-0.00	0.4030	-3.02	-0.11	2.94	-0.06	0.2016	0.6628	-0.0120	-0.0011	-0.0016	0.4030	0.6683	-0.0289	-0.0013	-0.0031	0.2016	0.6628
	8	-0.0134	0.0010	0.0017	0.4717	0.6960	-0.0235	-0.0013	-0.0031	0.2794											
	9																				
693	2	0.900	-0.003	-0.00	0.2903	-3.01	0.06	2.94	-0.02	0.2571	0.2848	-0.0135	-0.0007	-0.0015	0.2903	0.5537	0.0025	0.0000	0.0003	0.2571	0.5963
	8	-0.0135	0.0010	0.0015	0.3675	0.5525	0.0025	-0.0000	0.0003	-0.0887											
	9																				
693	3	0.900	0.51	-0.00	0.3614	-3.01	-0.05	2.94	-0.02	0.4652	0.6256	-0.0124	-0.0009	-0.0015	0.3614	0.6290	0.0045	0.0001	0.0009	0.4652	0.6878
	6	-0.0137	0.0010	0.0015	0.3770	0.5723	0.0066	0.0001	0.0009	0.0902											
	9																				
693	4	0.902	1.01	-0.00	0.3883	-3.01	0.04	2.95	-0.01	0.3184	0.6830	-0.0118	-0.0009	-0.0016	0.3883	0.6551	0.0094	0.0003	0.0012	0.3184	0.6784
	8	-0.0132	0.0010	0.0016	0.4068	0.6236	0.0104	0.0003	0.0014	0.1529											
	9																				
693	5	0.899	2.06	-0.00	0.3699	-3.01	-0.03	2.95	-0.00	0.2520	0.6619	-0.0125	-0.0009	-0.0016	0.3699	0.6653	0.0198	0.0006	0.0026	0.2520	0.6609
	8	-0.0135	0.0010	0.0016	0.4062	0.6168	0.0214	0.0006	0.0028	0.1541											
	9																				
693	6	0.898	3.08	-0.00	0.2893	-3.01	-0.01	2.95	-0.00	0.2353	0.6458	-0.0140	-0.0008	-0.0017	0.2893	0.6190	0.0305	0.0014	0.0039	0.2353	0.6443
	8	-0.0139	0.0011	0.0016	0.4035	0.6074	0.0318	0.0011	0.0041	0.1783											
	9																				

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

← See Key →

693	7	0.898	4.15	-0.0007	-0.0020	0.2496	-3.01	0.00	2.94	0.0051	0.2303	0.6422
	8	-0.0155	-0.0011	0.0017	-0.0017	0.3826	0.6429	0.0402	0.0018	0.0055	0.1871	0.6466
	9	-0.0149	0.0011	0.0000	0.0000	0.4190	0.6162	0.0021	0.0000	0.0002	1.1638	0.2145
693	7	0.897	-0.0009	-0.0010	-0.0016	0.3756	-3.01	-0.06	2.94	0.0000	-0.1106	0.5392
	8	-0.0123	-0.0009	-0.0016	-0.0016	0.4190	0.6162	0.0021	0.0000	0.0000	0.0000	0.6522
	9	-0.0131	0.0010	0.0000	0.0000	0.4260	0.7853	-0.0270	-0.0013	-0.0033	0.2710	0.6654
694	1	0.898	-3.09	-0.0000	-0.0006	0.0512	0.48	-0.11	-0.49	-0.07	0.2097	0.6522
	8	-0.0043	0.0000	0.0006	0.0006	0.4260	2.3092	-0.0253	-0.0013	-0.0033	0.2710	0.6654
	9	-0.0018	-0.0001	-0.0008	-0.0008	0.0961	0.9248	0.0007	-0.0000	0.0000	0.4904	0.5282
694	2	0.899	-0.0003	-0.0009	-0.0009	0.1510	0.48	-0.06	-0.50	0.03	-0.6092	0.1668
	8	-0.0053	0.0001	0.0006	0.0006	0.0961	1.0430	0.0022	0.0002	0.0002	0.4904	0.1668
	9	-0.0030	-0.0000	-0.0006	-0.0006	0.0486	0.7297	0.0007	0.0000	0.0006	0.4141	0.6810
694	3	0.897	0.49	-0.0000	-0.0005	0.0486	0.48	-0.05	-0.49	-0.02	0.4141	0.6810
	8	-0.0063	0.0000	0.0005	0.0005	0.0600	0.9052	0.0057	0.0004	0.0007	0.0813	0.6498
	9	-0.0032	-0.0000	-0.0005	-0.0005	0.433	0.7159	0.0048	0.0000	0.0006	0.2500	0.6556
694	4	0.898	1.01	-0.0000	-0.0005	0.0433	0.48	-0.05	-0.49	-0.01	0.2500	0.6556
	8	-0.0066	0.0000	0.0005	0.0005	0.0902	0.7159	0.0118	0.0005	0.0015	0.1600	0.6439
	9	-0.0030	-0.0000	-0.0005	-0.0005	0.0885	0.9794	0.0089	0.0002	0.0011	0.2360	0.6563
694	5	0.899	2.04	-0.0000	-0.0005	0.0885	0.48	-0.03	-0.49	-0.00	0.2360	0.6563
	8	-0.0060	0.0000	0.0005	0.0005	-0.0206	0.7154	0.0219	0.0010	0.0028	0.1597	0.6439
	9	-0.0040	0.0000	-0.0005	-0.0005	0.164	0.6245	0.0196	0.0006	0.0025	0.2213	0.6464
694	6	0.897	3.07	-0.0000	-0.0004	0.0164	0.49	-0.02	-0.49	0.01	0.2213	0.6464
	8	-0.0059	0.0000	0.0004	0.0004	-0.0393	0.6791	0.0293	0.0014	0.0037	0.1899	0.6369
	9	-0.0039	0.0000	-0.0004	-0.0004	0.0561	0.6238	0.0293	0.0011	0.0037	0.2192	0.6369
694	7	0.897	4.10	-0.0000	-0.0005	0.0561	0.48	-0.00	-0.48	0.02	0.2192	0.6408
	8	-0.0051	0.0000	0.0005	0.0005	-0.0221	0.6989	0.0425	0.0018	0.0054	0.1969	0.6369
	9	-0.0025	0.0000	-0.0005	-0.0005	0.0836	1.0087	0.0405	0.0015	0.0051	0.4067	0.4814
694	8	0.898	-0.0003	-0.0009	-0.0009	0.0836	0.48	-0.07	-0.49	-0.03	-0.4067	0.1159
	8	-0.0060	0.0001	0.0009	0.0009	-0.0352	0.7755	0.0024	0.0000	0.0002	0.9690	-0.1159
	9	-0.0038	0.0000	-0.0005	-0.0005	0.307	0.6717	0.0004	-0.0000	-0.0000	0.2184	0.6706
695	1	0.897	-3.10	-0.0000	-0.0004	0.307	0.48	-0.56	-0.49	0.42	0.2184	0.6706
	8	-0.0041	0.0000	0.0004	0.0004	-0.0637	0.7630	-0.0299	-0.0013	-0.0039	0.2184	0.6706
	9	-0.0043	0.0000	-0.0004	-0.0004	0.0637	0.5699	-0.0225	-0.0012	-0.0030	0.2184	0.6706

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key															
695	2	7 8 9	0.898 0.0059 -0.0036	-0.003 0.0000 0.0000	-0.00 0.0008 -0.0004	0.0417 -0.0216	0.48 0.7418 0.6765	-0.51 0.0001 0.0035	-0.48 0.0000 0.0000	0.47 -0.0000 0.0004	2.3752 0.0194	-1.2735 0.5743						
695	3	7 8 9	0.897 0.0056 -0.0028	0.50 0.0001 -0.0000	-0.00 0.0009 -0.0005	0.0962 0.0516	0.48 0.8283 0.9815	-0.49 0.0038 0.0072	-0.48 0.0003 0.0002	0.47 0.0004 0.0009	0.4227 0.1380	0.6315 0.6397						
695	4	7 8 9	0.898 0.0059 -0.0027	0.99 0.0000 -0.0000	-0.00 0.0009 -0.0005	0.0665 0.0893	0.48 0.7886 1.0435	-0.49 0.0081 0.0119	-0.49 0.0005 0.0003	0.48 0.0010 0.0015	0.3399 0.1629	0.6612 0.6274						
695	5	7 8 9	0.897 0.0062 -0.0031	2.02 0.0000 0.0000	-0.00 0.0008 -0.0005	0.0571 -0.0050	0.48 0.7099 0.8333	-0.47 0.0193 0.0223	-0.48 0.0008 0.0007	0.49 0.0025 0.0028	0.2312 0.1765	0.6593 0.6450						
695	6	7 8 9	0.898 0.0048 -0.0028	3.09 0.0001 -0.0000	-0.00 0.0008 -0.0005	0.1222 0.0037	0.48 0.8459 0.8935	-0.46 0.0293 0.0327	-0.48 0.0013 0.0012	0.50 0.0037 0.0041	0.2250 0.1929	0.6460 0.6353						
695	7	7 8 9	0.898 0.0043 -0.0028	4.13 0.0000 0.0000	-0.00 0.0006 -0.0004	0.0764 -0.0782	0.48 0.7387 0.7564	-0.44 0.0398 0.0443	-0.49 0.0017 0.0016	0.52 0.0051 0.0056	0.2167 0.1917	0.6438 0.6337						
695	8	7 8 9	0.898 0.0061 -0.0037	-0.02 0.0000 0.0000	-0.00 0.0008 -0.0005	-0.0136 -0.0406	0.49 0.6767 0.6824	-0.50 0.0002 0.0035	-0.48 0.0000 0.0000	0.46 -0.0000 0.0004	2.2255 0.0579	-0.3011 0.6102						
696	1	7 8 9	0.899 0.0075 -0.0069	-3.11 0.0001 -0.0001	-0.00 0.0009 -0.0008	0.1089 0.0954	1.02 0.6631 0.6375	-0.11 -0.0271 -0.0259	-1.03 -0.0011 -0.0013	-0.07 -0.0035 -0.0034	0.2080 0.2664	0.6474 0.6659						
696	2	7 8 9	0.897 0.0090 -0.0068	-0.02 0.0002 -0.0001	-0.00 0.0013 -0.0008	0.1378 0.0866	1.02 0.7367 0.6010	-0.05 0.0026 0.0006	-1.03 0.0002 -0.0000	-0.03 0.0000 0.0000	0.4192 -0.6605	0.5514 0.1332						
696	3	7 8 9	0.898 0.0098 -0.0056	0.51 0.0001 -0.0001	-0.00 0.0013 -0.0009	0.0867 0.1585	1.02 0.6726 0.8129	-0.05 0.0062 0.0045	-1.03 0.0004 0.0000	-0.02 0.0008 0.0005	0.3692 0.0915	0.6683 0.6348						
696	4	7 8 9	0.900 0.0095 -0.0051	1.02 0.0002 -0.0002	-0.00 0.0014 -0.0009	0.1339 0.2093	1.02 0.7505 0.9458	-0.04 0.0124 0.0086	-1.03 0.0005 0.0002	-0.02 0.0016 0.0011	0.2343 0.1599	0.6701 0.6609						

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

See Key

Run Pt.	Cd																			
696	5	7	0.900	2.04	-0.00	1.02	-0.023	-1.03	0.00	0.2283	0.6570									
	8	8	0.0097	0.0002	0.0013	0.7166	0.0221	0.0010	0.0029	0.1608	0.6550									
	9	9	-0.0060	-0.0001	-0.0008	0.7114	0.0189	0.0006	0.0024											
696	6	7	0.900	3.10	-0.00	1.02	-0.02	-1.03	0.00	0.2170	0.6503									
	8	8	0.0091	0.0001	0.0013	0.7247	0.0330	0.0014	0.0042	0.1927	0.6455									
	9	9	-0.0060	-0.0001	-0.0007	0.6245	0.0293	0.0011	0.0037											
696	7	7	0.897	4.14	-0.00	1.02	-0.00	-1.02	0.02	0.2165	0.6426									
	8	8	0.0081	0.0001	0.0011	0.7025	0.0427	0.0018	0.0054	0.1969	0.6402									
	9	9	-0.0055	-0.0001	-0.0007	0.6432	0.0402	0.0015	0.0051											
696	8	7	0.898	-0.03	-0.00	1.02	-0.07	-1.03	-0.03	0.4454	0.5567									
	8	8	0.0094	0.0001	0.0012	0.6838	0.0025	0.0002	0.0002	-1.1719	-0.5311									
	9	9	-0.0068	-0.0000	-0.0008	0.5898	0.0003	-0.0000	-0.0000											
697	1	7	0.897	-3.11	-0.00	1.01	-1.00	-1.03	0.92	0.2234	0.6508									
	8	8	0.0061	0.0002	0.0009	0.7877	-0.0323	-0.0014	-0.0042	0.2645	0.6628									
	9	9	-0.0052	-0.0002	-0.0009	0.8913	-0.0198	-0.0010	-0.0026											
697	2	7	0.898	-0.02	-0.00	1.02	-0.94	-1.03	0.96	0.2252	0.9645									
	8	8	0.0087	0.0001	0.0011	0.6874	-0.0014	-0.0001	-0.0002	0.1705	0.7106									
	9	9	-0.0058	-0.0001	-0.0008	0.6966	0.0055	0.0001	-0.0002	0.4885	0.6784									
697	3	7	0.898	0.50	-0.00	1.02	-0.94	-1.03	0.97	0.4885	0.5973									
	8	8	0.0088	0.0002	0.0012	0.7253	0.0021	0.0003	0.0002	0.1873	0.6784									
	9	9	-0.0054	-0.0001	-0.0008	0.7530	0.0095	0.0003	0.0012											
697	4	7	0.897	1.02	-0.00	1.02	-0.93	-1.03	0.97	0.3684	0.6818									
	8	8	0.0096	0.0001	0.0012	0.6402	0.0061	0.0004	0.0006	0.1710	0.6674									
	9	9	-0.0050	-0.0001	-0.0008	0.8319	0.0151	0.0005	0.0020											
697	5	7	0.897	2.02	-0.00	1.02	-0.92	-1.03	0.98	0.2271	0.6657									
	8	8	0.0091	0.0002	0.0013	0.7477	0.0171	0.0007	0.0022	0.1875	0.6430									
	9	9	-0.0044	-0.0002	-0.0008	0.9557	0.0248	0.0009	0.0031											
697	6	7	0.898	3.09	-0.00	1.02	-0.90	-1.03	1.00	0.2213	0.6473									
	8	8	0.0084	0.0002	0.0011	0.7087	0.0273	0.0012	0.0035	0.1961	0.6408									
	9	9	-0.0050	-0.0001	-0.0007	0.7745	0.0356	0.0013	0.0045											
697	7	7	0.899	4.14	-0.00	1.02	-0.88	-1.03	1.01	0.2157	0.6461									
	8	8	0.0080	0.0001	0.0010	0.6472	0.0374	0.0016	0.0048	0.1948	0.6301									
	9	9	-0.0044	-0.0001	-0.0007	0.8628	0.0465	0.0018	0.0058											

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																
697	8	0.897	-0.0082	-0.0060	-0.0001	-0.0002	-0.0012	0.1519	0.0887	1.02	0.7478	0.6582	-0.0015	-0.0001	-1.03	0.96	0.1983	0.8976
	8	0.0082															0.1599	0.6021
	9	-0.0060																
698	1	0.898	-0.0131	-0.0137	-0.0003	-0.0005	-0.0018	0.2273	0.1442	2.08	0.6989	0.6595	-0.0261	-0.0013	-2.12	0.07	0.2051	0.6316
	8	0.0131															0.2601	0.6623
	9	-0.0137																
69A	2	0.899	-0.0160	-0.0130	-0.0004	-0.0020	-0.0016	0.1458	0.1657	2.08	0.6409	0.6238	-0.0029	-0.0000	-2.11	0.03	0.4433	0.6782
	8	0.0160															0.8249	0.7640
	9	-0.0130																
69B	3	0.899	-0.0161	-0.0123	-0.0004	-0.0021	-0.0016	0.1576	0.1926	2.09	0.6502	0.6742	-0.0067	-0.0000	-2.12	0.02	0.3476	0.6776
	8	0.0161															0.1170	0.7130
	9	-0.0123																
69B	4	0.895	-0.0167	-0.0116	-0.0005	-0.0021	-0.0017	0.1603	0.2223	2.09	0.6570	0.7354	-0.0132	-0.0002	-2.12	0.01	0.2305	0.6814
	8	0.0167															0.1785	0.6772
	9	-0.0116																
69B	5	0.899	-0.0167	-0.0115	-0.0005	-0.0022	-0.0016	0.1512	0.2197	2.09	0.6628	0.7075	-0.0230	-0.0006	-2.12	0.00	0.2271	0.6663
	8	0.0167															0.1691	0.6596
	9	-0.0115																
69B	6	0.898	-0.0163	-0.0125	-0.0004	-0.0021	-0.0014	0.1267	0.1628	2.08	0.6643	0.5759	-0.0341	-0.0011	-2.11	0.00	0.2156	0.6537
	8	0.0163															0.1978	0.6453
	9	-0.0125																
69A	7	0.898	-0.0157	-0.0114	-0.0004	-0.0021	-0.0014	0.1081	0.1786	2.08	0.6711	0.6223	-0.0434	-0.0016	-2.11	0.02	0.2131	0.6465
	8	0.0157															0.2009	0.6422
	9	-0.0114																
69B	8	0.898	-0.0152	-0.0127	-0.0004	-0.0021	-0.0016	0.1904	0.1791	2.09	0.6924	0.6656	-0.0030	-0.0000	-2.12	0.03	0.4031	0.6268
	8	0.0152															2.0507	2.5819
	9	-0.0127																
699	1	0.898	-0.0126	-0.0120	-0.0004	-0.0016	-0.0016	0.1932	0.1646	2.08	0.6418	0.6873	-0.0370	-0.0007	-2.11	0.94	0.2272	0.6505
	8	0.0126															0.2661	0.6885
	9	-0.0120																
699	2	0.900	-0.0152	-0.0111	-0.0004	-0.0018	-0.0015	0.1308	0.2033	2.08	0.6234	0.7026	-0.0057	-0.0004	-2.11	1.97	0.2384	0.7203
	8	0.0152															0.2120	0.6934
	9	-0.0111																

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																
699	3	7	0.899	0.49	-0.00	0.1404	2.08	-1.84	-2.11	1.98	0.2761	0.8755	0.0158	0.0013	-0.0000	-0.0002	0.1743	0.6708	
		8	0.0158	0.0004	0.0019	0.2089	0.6273	-0.0158	0.0005	0.0002									
		9	-0.0108	-0.0004	-0.0015	0.2089	0.7032	0.0158	0.0005	0.0021									
699	4	7	0.900	1.01	-0.00	0.1684	2.08	-1.84	-2.11	1.98	0.3703	0.5835	0.0153	0.0026	0.0001	0.0003	0.1868	0.6673	
		8	0.0153	0.0005	0.0020	0.2068	0.6477	0.0026	0.0007	0.0027									
		9	-0.0110	-0.0004	-0.0014	0.2068	0.6767	0.0206	0.0007	0.0027									
699	5	7	0.898	2.06	-0.00	0.1508	2.09	-1.82	-2.11	1.99	0.2280	0.6558	0.0156	0.0121	0.0005	0.0015	0.2060	0.6583	
		8	0.0156	0.0004	0.0020	0.2264	0.6566	0.0121	0.0005	0.0015									
		9	-0.0104	-0.0004	-0.0015	0.2264	0.7171	0.0305	0.0012	0.0040									
699	6	7	0.898	3.12	-0.00	0.1619	2.09	-1.81	-2.11	2.01	0.2116	0.6623	0.0146	0.0218	0.0009	0.0028	0.2034	0.6455	
		8	0.0146	0.0004	0.0020	0.2145	0.6914	0.0218	0.0009	0.0028									
		9	-0.0100	-0.0004	-0.0014	0.2145	0.6943	0.0414	0.0016	0.0053									
699	7	7	0.899	4.12	-0.00	0.1374	2.09	-1.79	-2.11	2.03	0.2021	0.6439	0.0145	0.0325	0.0013	0.0041	0.2016	0.6251	
		8	0.0145	0.0004	0.0019	0.2102	0.6845	0.0325	0.0020	0.0064									
		9	-0.0097	-0.0004	-0.0013	0.2102	0.6801	0.0513	0.0020	0.0064									
699	8	7	0.899	-2.12	-0.00	0.1940	2.08	-1.88	-2.11	1.95	0.2234	0.6513	0.0129	-0.0270	-0.0012	-0.0035	0.2178	0.7620	
		8	0.0129	0.0005	0.0017	0.1807	0.6642	-0.0270	-0.0003	-0.0008									
		9	-0.0116	-0.0004	-0.0016	0.1807	0.7197	-0.0058	-0.0003	-0.0008									
699	9	7	0.900	-0.02	-0.00	0.1543	2.09	-1.85	-2.11	1.97	0.2298	0.6890	0.0149	-0.0059	-0.0002	-0.0008	0.2120	0.7154	
		8	0.0149	0.0004	0.0019	0.1845	0.6343	-0.0059	-0.0004	0.0014									
		9	-0.0114	-0.0004	-0.0015	0.1845	0.6685	0.0103	-0.0004	0.0014									
700	1	7	0.899	-3.09	-0.00	0.2414	5.06	-0.10	-5.06	-0.07	0.2229	0.6491	0.0310	-0.0242	-0.0010	-0.0031	0.2378	0.6667	
		8	0.0310	0.0015	0.0040	0.1786	0.6552	-0.0242	-0.0013	-0.0038									
		9	-0.0330	-0.0011	-0.0043	0.1786	0.6552	-0.0285	-0.0013	-0.0038									
700	2	7	0.899	-0.06	-0.00	0.2229	5.06	-0.06	-5.05	-0.03	0.3520	0.6961	0.0338	-0.0041	-0.0002	-0.0005	0.3308	0.8269	
		8	0.0338	0.0015	0.0044	0.2463	0.6624	0.0041	0.0002	0.0005									
		9	-0.0296	-0.0014	-0.0039	0.2463	0.6664	-0.0019	-0.0001	-0.0003									
700	3	7	0.901	0.50	-0.00	0.2014	5.06	-0.05	-5.05	-0.03	0.2875	0.6913	0.0346	-0.0087	0.0005	0.0012	0.1073	0.5973	
		8	0.0346	0.0013	0.0044	0.2497	0.6435	0.0087	0.0005	0.0012									
		9	-0.0295	-0.0014	-0.0038	0.2497	0.6555	0.0024	0.0000	0.0002									
700	4	7	0.901	0.98	-0.00	0.2065	5.06	-0.04	-5.06	-0.02	0.1938	0.6833	0.0348	-0.0150	0.0005	0.0020	0.1978	0.6662	
		8	0.0348	0.0014	0.0045	0.2517	0.6399	0.0150	0.0005	0.0020									
		9	-0.0294	-0.0014	-0.0038	0.2517	0.6511	0.0062	0.0002	0.0008									

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key												
700	5	7	0.899	2.06	0.00	0.1985	5.06	-0.03	-5.05	-0.00	0.2012	0.6654			
		8	0.353	0.0014	0.0047	0.2686	0.6751	0.251	0.0010	0.0033	0.1725	0.6517			
		9	-0.0282	-0.0015	-0.0038		0.6769	0.0163	0.0005	0.0021					
700	6	7	0.900	3.11	0.00	0.1639	5.06	-0.00	-5.06	0.00	0.1940	0.6564			
		8	0.361	0.0011	0.0048	0.2505	0.6664	0.360	0.0013	0.0047	0.2044	0.6488			
		9	-0.0286	-0.0014	-0.0035		0.6264	0.0266	0.0010	0.0034					
700	7	7	0.898	4.14	0.00	0.1460	5.06	-0.00	-5.05	0.02	0.1990	0.6502			
		8	0.358	0.0010	0.0048	0.2586	0.6730	0.455	0.0018	0.0059	0.2098	0.6407			
		9	-0.0277	-0.0014	-0.0035		0.6423	0.0376	0.0015	0.0048					
700	8	7	0.901	-0.01	-0.00	0.2269	5.06	-0.06	-5.05	-0.03	0.3374	0.6901			
		8	0.336	0.0015	0.0044	0.2362	0.6638	0.0044	0.0003	0.0006	0.3256	0.6634			
		9	-0.0300	-0.0014	-0.0038		0.6475	-0.0018	-0.0001	-0.0003					
701	1	7	0.899	-3.10	-0.00	0.2405	5.06	-5.11	-5.05	5.02	0.2262	0.6167			
		8	0.280	0.0013	0.0035	0.1899	0.6419	-0.0532	-0.0024	-0.0065	0.4415	1.2990			
		9	-0.0307	-0.0011	-0.0038		0.6304	0.0002	0.0000	0.0000					
701	2	7	0.900	-0.04	-0.00	0.2310	5.06	-5.07	-5.05	5.06	0.2376	0.6749			
		8	0.308	0.0014	0.0040	0.2674	0.6556	-0.0232	-0.0011	-0.0031	0.2174	0.6724			
		9	-0.0264	-0.0014	-0.0035		0.6784	0.0278	0.0012	0.0037					
701	3	7	0.898	0.52	0.00	0.2334	5.06	-5.06	-5.05	5.06	0.2307	0.6851			
		8	0.311	0.0014	0.0041	0.2600	0.6595	-0.0179	-0.0008	-0.0024	0.2164	0.6584			
		9	-0.0263	-0.0013	-0.0034		0.6609	0.0328	0.0014	0.0043					
701	4	7	0.896	1.01	-0.00	0.2328	5.06	-5.05	-5.05	5.07	0.2348	0.7145			
		8	0.316	0.0014	0.0041	0.2537	0.6582	-0.0130	-0.0006	-0.0018	0.2180	0.6580			
		9	-0.0264	-0.0013	-0.0033		0.6372	0.0376	0.0016	0.0049					
701	5	7	0.897	2.05	0.00	0.2190	5.06	-5.03	-5.05	5.09	0.4745	0.8560			
		8	0.323	0.0014	0.0043	0.2697	0.6664	-0.0028	-0.0002	-0.0004	0.2111	0.6312			
		9	-0.0255	-0.0013	-0.0033		0.6636	0.0474	0.0020	0.0059					
701	6	7	0.896	3.08	0.00	0.1797	5.06	-5.02	-5.05	5.10	0.1822	0.6398			
		8	0.331	0.0011	0.0042	0.3174	0.6456	0.0046	0.0001	0.0005	0.2132	0.6100			
		9	-0.0230	-0.0014	-0.0034		0.7544	0.0555	0.0023	0.0067					
701	7	7	0.898	4.14	0.00	0.1654	5.05	-5.00	-5.05	5.11	0.1623	0.6442			
		8	0.330	0.0010	0.0043	0.2628	0.6655	0.0141	0.0004	0.0018	0.2184	0.6442			
		9	-0.0242	-0.0012	-0.0031		0.6450	0.0633	0.0027	0.0075					

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key															
701	8	0.899	-0.0015	-0.0014	-0.003	0.0040	0.2503	0.6713	-0.0228	-5.06	-5.05	-0.0011	5.06	0.2410	0.6780	0.6336	0.6336
	9	-0.0262	0.0014	-0.0035	0.0040	0.2695	0.6776	0.6776	0.0280	-0.0012	0.0037	0.0037	0.2187	0.6598	0.6574		
702	3	0.899	-0.0020	-0.0044	4.99	0.2862	0.6095	-3.05	-3.00	-3.05	-0.0017	-3.01	0.2438	0.6598	0.6574		
	9	-0.0426	0.0015	-0.0055	4.99	0.1852	0.6484	0.6484	-0.0377	-0.0015	-0.0049	-0.0049	0.2073	0.6598	0.6574		
702	4	0.897	-0.0017	-0.0024	4.99	0.3660	0.6262	-3.01	-2.96	-3.01	-0.0006	-2.99	0.2004	0.7013	0.7083		
	9	-0.0152	0.0008	-0.0024	4.99	0.2304	0.6973	0.6973	-0.0142	-0.0009	-0.0020	-0.0020	0.3254	0.7013	0.7083		
702	5	0.899	-0.0006	-0.0015	4.99	0.3813	0.5444	-3.00	-2.95	-3.00	-0.0003	-2.98	0.1649	0.7457	0.7600		
	9	-0.0094	0.0004	-0.0015	4.99	0.2626	0.8051	0.8051	-0.0080	-0.0006	-0.0012	-0.0012	0.4007	0.7457	0.7600		
702	6	0.898	-0.0003	-0.0003	4.99	-0.1181	0.8906	-2.98	-2.93	-2.97	-0.0001	-2.96	0.5882	0.7866	0.8484		
	9	0.0041	0.0003	0.0001	4.99	1.1278	0.4461	0.4461	0.0015	0.0001	0.0001	0.0001	-0.3195	0.7866	0.8484		
702	7	0.898	0.0014	0.0027	4.99	0.1553	0.6529	-2.94	-2.89	-2.93	0.0007	-2.93	0.1518	0.6303	0.6196		
	9	0.0253	0.0014	0.0027	4.99	0.3126	0.5988	0.5988	0.0211	0.0007	0.0026	0.0026	0.1832	0.6303	0.6196		
702	8	0.900	0.0024	0.0051	4.99	0.1667	0.6239	-2.91	-2.86	-2.89	0.0014	-2.89	0.1595	0.6291	0.6243		
	9	0.0465	0.0024	0.0051	4.99	0.2975	0.6532	0.6532	0.0439	0.0020	0.0054	0.0054	0.2282	0.6291	0.6243		
702	9	0.899	0.0020	0.0068	4.99	0.1441	0.6527	-2.88	-2.83	-2.86	0.0019	-2.87	0.1461	0.6153	0.6153		
	9	0.0697	0.0020	0.0068	4.99	0.2867	0.5847	0.5847	0.0614	0.0028	0.0071	0.0071	0.2302	0.6153	0.6153		
702	10	0.899	-0.0007	-0.0024	4.99	0.3097	0.5806	-3.01	-2.95	-3.01	-0.0006	-2.99	0.2114	0.7185	0.7465		
	9	-0.0159	0.0007	-0.0024	4.99	0.2199	0.7131	0.7131	-0.0141	-0.0009	-0.0021	-0.0021	0.3311	0.7185	0.7465		
703	1	0.897	-0.0011	-0.0021	4.99	0.3191	0.5984	-0.06	-0.09	0.03	-0.0008	-0.06	0.2229	0.6821	0.7033		
	9	-0.0180	0.0011	-0.0021	4.99	0.1520	0.6226	0.6226	-0.0172	-0.0008	-0.0024	-0.0024	0.2603	0.6821	0.7033		
703	2	0.897	-0.0001	-0.0001	4.99	-1.0615	0.04	-0.04	-0.06	0.07	-0.0001	-0.03	2.6600	0.6243	0.2099		
	9	0.0008	0.0002	-0.0001	4.99	-2.1848	1.5324	1.5324	0.0009	-0.0009	-0.0001	-0.0001	-0.3666	0.6243	0.2099		

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																			
703	3	7 8 9	0.899	0.98	44.99	0.1315	-0.01	-0.05	0.08	-0.02	0.4294	0.5660	0.0074	0.0001	0.0012	0.0045	0.0003	0.0005	0.1110	0.6479		
			0.0038	0.0006	0.0007	0.8628	0.9196	0.0068	0.0001	0.0008												
			0.900	3.07	44.99	0.1881	0.6813	-0.03	0.11	-0.01	0.1866	0.6491	0.0233	0.0008	0.0013	0.0023	0.0191	0.0007	0.0024	0.1838	0.6539	
703	5	7 8 9	0.900	6.18	44.99	0.1700	0.6712	-0.00	0.15	0.02	0.1482	0.6439	0.0458	0.0023	0.0061	0.0435	0.0012	0.0056	0.2201	0.6296		
			0.0416	0.0023	0.0048	0.2832	0.5864	0.0394	0.0016	0.0049												
			0.899	9.30	44.99	0.1470	0.6375	0.02	0.18	0.04	0.1508	0.6166	0.0673	0.0032	0.0019	0.0085	0.0640	0.0027	0.0079	0.2355	0.5891	
703	7	7 8 9	0.897	12.43	44.99	0.1326	0.6166	0.04	0.20	0.07	0.1349	0.6139	0.0864	0.0037	0.0106	0.0843	0.0022	0.0103	0.2246	0.5626		
			0.0708	0.0037	0.0079	0.2633	0.5635	0.0724	0.0032	0.0081												
			0.898	-0.02	44.99	-0.9806	-0.03	-0.06	0.07	-0.03	0.9405	-3.1757	0.0012	0.0003	0.0002	0.0000	0.0002	0.0000	0.0000	-0.5248	-0.0071	
704	1	7 8 9	0.900	-3.12	44.99	0.3061	0.5303	0.98	0.98	0.93	0.1972	0.7317	-0.0119	0.0004	0.0013	-0.0123	0.0004	0.0016	0.2879	0.7293		
			0.0149	0.0004	0.0020	0.1500	0.6688	-0.0113	0.0006	0.0016												
			0.898	-0.01	44.99	0.0979	0.8251	1.01	1.02	0.96	0.3977	0.5526	0.0063	0.0001	0.0005	0.0008	0.0051	0.0001	0.0008	0.1173	0.6584	
704	3	7 8 9	0.899	1.00	44.99	0.1791	0.7323	1.02	1.03	0.97	0.2349	0.6351	0.0144	0.0009	0.0021	0.0121	0.0004	0.0015	0.1695	0.6403		
			0.0088	0.0009	0.0013	0.5340	0.7479	0.0126	0.0004	0.0016												
			0.900	3.08	44.99	0.1559	0.6420	1.04	1.06	0.99	0.1746	0.6460	0.0316	0.0009	0.0016	0.0033	0.0267	0.0010	0.0032	0.2066	0.6471	
704	5	7 8 9	0.899	6.18	44.99	0.1520	0.6439	1.08	1.10	1.02	0.1522	0.6392	0.0537	0.0016	0.0069	0.0510	0.0015	0.0065	0.2314	0.6272		
			0.0439	0.0027	0.0056	0.3173	0.6467	0.0455	0.0021	0.0057												
			0.900	9.30	44.99	0.1315	0.6166	0.04	0.07	0.04	0.1508	0.6166	0.0673	0.0032	0.0019	0.0085	0.0640	0.0027	0.0079	0.2355	0.5891	

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key										
704	6	7	8	9	44.99	0.1433	1.11	1.10	1.13	1.05	0.1480	0.6107
					0.0090	0.2966	0.225	0.0701	0.0020	0.0085	0.2348	0.5808
					0.0071	0.5982	0.5982	0.0623	0.0029	0.0072		
704	7	7	8	9	44.99	0.1348	1.13	1.12	1.15	1.07	0.1319	0.6091
					0.0112	0.2569	0.6185	0.0901	0.0023	0.0109	0.2181	0.5594
					0.0079	0.5467	0.5467	0.0756	0.0033	0.0084		
704	8	7	8	9	44.99	0.0987	1.01	1.01	1.02	0.96	0.3641	0.5164
					0.0010	0.9267	0.7984	0.0053	0.0003	0.0005	0.1195	0.6154
					0.0004	0.6943	0.6943	0.0063	0.0001	0.0007		
705	1	7	8	9	44.99	1.1127	3.01	2.97	2.90	3.04	-0.3426	1.3003
					0.0000	-0.0950	-0.7966	-0.0003	-0.0000	-0.0002	-1.2863	1.6188
					-0.0034							
705	2	7	8	9	44.99	0.2206	3.04	3.00	2.94	3.07	0.2401	0.6543
					0.0027	0.4592	0.7167	0.0179	0.0008	0.0023	0.1825	0.6686
					0.0019	0.7330	0.7330	0.0190	0.0006	0.0025		
705	3	7	8	9	44.99	0.2055	3.06	3.01	2.95	3.07	0.2197	0.6546
					0.0038	0.3296	0.6512	0.0256	0.0010	0.0033	0.2057	0.6541
					0.0029							
705	4	7	8	9	44.99	0.1714	3.08	3.03	2.98	3.09	0.1699	0.6588
					0.0060	0.3239	0.6645	0.0412	0.0014	0.0054	0.2194	0.6407
					0.0047	0.6613	0.6613	0.0385	0.0016	0.0049		
705	5	7	8	9	44.99	0.1606	3.11	3.06	3.02	3.12	0.1591	0.6138
					0.0082	0.2871	0.5896	0.0554	0.0025	0.0067	0.2342	0.6044
					0.0031							
705	6	7	8	9	44.99	0.1436	3.13	3.09	3.04	3.15	0.1451	0.6047
					0.0101	0.2652	0.6132	0.0807	0.0023	0.0097	0.2303	0.5647
					0.0078	0.5712	0.5712	0.0704	0.0032	0.0079		
705	7	7	8	9	44.99	0.1325	3.15	3.11	3.04	3.16	0.1302	0.6027
					0.0122	0.2286	0.6122	0.0806	0.0025	0.0118	0.1979	0.5472
					0.0085	0.5441	0.5441	0.0806	0.0031	0.0088		
705	8	7	8	9	44.99	0.1737	3.04	3.00	2.94	3.07	0.2320	0.6404
					0.0026	0.3601	0.6377	0.0180	0.0007	0.0023	0.1830	0.6544
					0.0017	0.5658	0.5658	0.0191	0.0007	0.0025		

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key													
706	1	7	0.900	-3.008	44.99	0.1960	0.604	6.02	6.06	6.06	0.0021	0.2779	0.6748		
		6	0.0171	0.0006	0.0022	0.5253	0.6539	0.0162	0.0009	0.0005	0.0024	0.1601	0.6780		
		9	0.0111	0.0011	0.0017	0.0000	0.7799	0.0177	0.0005	0.0005	0.0000	0.0000	0.0000		
706	2	7	0.900	0.02	44.99	0.2063	0.607	6.05	6.11	6.04	0.0051	0.2097	0.6575		
		6	0.0398	0.0016	0.0054	0.3073	0.6845	0.0387	0.0016	0.0016	0.0049	0.2125	0.6573		
		9	0.0344	0.0021	0.0046	0.0000	0.6676	0.0378	0.0016	0.0016	0.0000	0.0000	0.0000		
706	3	7	0.899	1.05	44.99	0.1802	0.608	6.06	6.12	6.05	0.0059	0.1935	0.6470		
		6	0.0490	0.0017	0.0064	0.2919	0.6588	0.0460	0.0017	0.0019	0.0057	0.2151	0.6458		
		9	0.0413	0.0024	0.0051	0.0000	0.6258	0.0443	0.0019	0.0019	0.0000	0.0000	0.0000		
706	4	7	0.898	3.14	44.99	0.1797	0.610	6.07	6.15	6.06	0.0071	0.1787	0.6182		
		6	0.0615	0.0022	0.0077	0.2846	0.6316	0.0578	0.0020	0.0024	0.0068	0.2192	0.6199		
		9	0.0539	0.0030	0.0064	0.0000	0.5987	0.0549	0.0024	0.0024	0.0000	0.0000	0.0000		
706	5	7	0.897	6.22	44.99	0.1547	0.613	6.10	6.18	6.10	0.0092	0.1591	0.5994		
		6	0.0791	0.0024	0.0096	0.2703	0.6089	0.0768	0.0024	0.0031	0.0078	0.2289	0.5719		
		9	0.0679	0.0036	0.0078	0.0000	0.5783	0.0684	0.0031	0.0031	0.0000	0.0000	0.0000		
706	6	7	0.898	9.35	44.99	0.1435	0.614	6.12	6.18	6.10	0.0111	0.1445	0.5924		
		6	0.0957	0.0027	0.0115	0.2366	0.6031	0.0940	0.0027	0.0033	0.0130	0.2054	0.5539		
		9	0.0786	0.0037	0.0086	0.0000	0.5504	0.0814	0.0033	0.0033	0.0090	0.1637	0.5907		
706	7	7	0.899	12.47	44.99	0.1322	0.616	6.14	6.18	6.10	0.0130	0.1295	0.5907		
		6	0.1112	0.0029	0.0132	0.1824	0.5949	0.1105	0.0028	0.0029	0.0130	0.2066	0.6470		
		9	0.0862	0.0031	0.0095	0.0000	0.5518	0.0902	0.0029	0.0029	0.0099	0.2125	0.6514		
706	8	7	0.898	0.03	44.99	0.2037	0.608	6.05	6.11	6.04	0.0050	0.2066	0.6470		
		6	0.0401	0.0016	0.0054	0.3066	0.6779	0.0391	0.0016	0.0016	0.0049	0.2125	0.6514		
		9	0.0346	0.0021	0.0045	0.0000	0.6600	0.0379	0.0016	0.0016	0.0049	0.2125	0.6514		
707	1	7	0.898	-3.04	44.99	0.2436	0.907	9.11	9.04	9.02	0.0047	0.2458	0.6569		
		6	0.0363	0.0017	0.0049	0.2769	0.6829	0.0362	0.0017	0.0013	0.0047	0.1885	0.6653		
		9	0.0332	0.0018	0.0043	0.0000	0.6568	0.0365	0.0013	0.0013	0.0047	0.2042	0.6240		
707	2	7	0.900	0.04	44.99	0.2097	0.910	9.14	9.08	9.05	0.0068	0.2117	0.6172		
		6	0.0564	0.0023	0.0073	0.2719	0.6481	0.0550	0.0023	0.0022	0.0068	0.2042	0.6240		
		9	0.0512	0.0027	0.0061	0.0000	0.6034	0.0544	0.0022	0.0022	0.0068	0.2042	0.6240		
707	3	7	0.899	1.08	44.99	0.1948	0.911	9.14	9.09	9.06	0.0074	0.1959	0.6075		
		6	0.0629	0.0024	0.0078	0.2744	0.6232	0.0610	0.0023	0.0024	0.0074	0.2081	0.6075		
		9	0.0566	0.0031	0.0068	0.0000	0.6055	0.0597	0.0024	0.0024	0.0074	0.2081	0.6075		

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key														
707	4	7	0.900	3.15	44.99	0.1792	0.6107	9.12	9.15	9.10	9.07	0.1753	0.5955				
		8	0.0746	0.0026	0.0091	0.2564	0.5534	0.0725	0.0025	0.0078	0.2194	0.5827					
		9	0.0668	0.0034	0.0073			0.0673	0.0029	0.0078							
707	5	7	0.899	6.24	44.99	0.1577	0.6007	9.17	9.12	9.08	0.1565	0.5890					
		8	0.0916	0.0028	0.0110	0.2361	0.5563	0.0898	0.0028	0.0105	0.2060	0.5602					
		9	0.0796	0.0037	0.0088			0.0781	0.0032	0.0087							
707	6	7	0.899	9.36	44.99	0.1335	0.5824	9.19	9.12	9.09	0.1399	0.5883					
		8	0.1083	0.0028	0.0126	0.1962	0.5637	0.1076	0.0030	0.0126	0.1746	0.5528					
		9	0.0866	0.0034	0.0097			0.0894	0.0031	0.0098							
707	7	7	0.899	12.48	44.99	0.1327	0.5950	9.20	9.12	9.08	0.1258	0.5852					
		8	0.1214	0.0032	0.0144	0.1588	0.5460	0.1219	0.0030	0.0142	0.1404	0.5539					
		9	0.0937	0.0029	0.0102			0.0944	0.0026	0.0104							
707	8	7	0.899	0.04	44.99	0.1998	0.6296	9.14	9.07	9.05	0.2081	0.6105					
		8	0.0565	0.0022	0.0071	0.2879	0.6265	0.0552	0.0022	0.0067	0.2027	0.6188					
		9	0.0501	0.0028	0.0062			0.0544	0.0022	0.0067							
708	1	7	0.901	-2.99	44.99	0.2191	0.5884	15.11	15.11	15.09	0.2201	0.5953					
		8	0.0681	0.0029	0.0084	0.2314	0.5884	0.0665	0.0029	0.0082	0.1836	0.6049					
		9	0.0659	0.0030	0.0077			0.0684	0.0025	0.0082							
708	2	7	0.900	0.06	44.99	0.1877	0.5714	15.12	15.13	15.10	0.1960	0.5839					
		8	0.0832	0.0031	0.0098	0.2234	0.5896	0.0819	0.0032	0.0095	0.1856	0.5744					
		9	0.0794	0.0035	0.0090			0.0811	0.0030	0.0093							
708	3	7	0.900	1.10	44.99	0.1731	0.5934	15.12	15.14	15.11	0.1837	0.5835					
		8	0.0900	0.0031	0.0106	0.2147	0.5719	0.0874	0.0032	0.0096	0.1833	0.5697					
		9	0.0840	0.0036	0.0096			0.0847	0.0031	0.0096							
708	4	7	0.898	3.17	44.99	0.1533	0.5833	15.13	15.15	15.12	0.1634	0.5805					
		8	0.0992	0.0030	0.0115	0.2002	0.5589	0.0974	0.0031	0.0113	0.1758	0.5635					
		9	0.0904	0.0036	0.0101			0.0910	0.0032	0.0102							
708	5	7	0.901	6.27	44.99	0.1423	0.5935	15.14	15.15	15.11	0.1307	0.5869					
		8	0.1124	0.0032	0.0133	0.1570	0.5522	0.1138	0.0029	0.0133	0.1507	0.5577					
		9	0.0973	0.0030	0.0107			0.0984	0.0029	0.0109							
708	6	7	0.899	9.38	44.99	0.1262	0.5906	15.15	15.15	15.12	0.1182	0.5828					
		8	0.1263	0.0031	0.0149	0.1451	0.5531	0.1271	0.0030	0.0148	0.1252	0.5587					
		9	0.1019	0.0029	0.0112			0.1036	0.0025	0.0115							

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key															
708	7	0.900	12.50	44.99	0.1047	15.19	15.16	15.15	15.12	0.1041	0.5812	0.1387	0.0027	0.0161	0.1251	0.5812	0.5527
	8	0.1374	0.0028	0.0159	0.1378	0.5598	0.1083	0.0028	0.0119								
	9	0.1044	0.0028	0.0117	0.1378	0.5598	0.1083	0.0027	0.0119								
708	7	0.901	0.08	44.99	0.1850	15.14	15.12	15.14	15.10	0.1930	0.5934	0.0823	0.0031	0.0095	0.1835	0.5813	0.5685
	8	0.0835	0.0030	0.0099	0.2182	0.5612	0.0813	0.0029	0.0092								
	9	0.0802	0.0035	0.0090	0.2182	0.5612	0.0813	0.0029	0.0092								
709	3	0.898	-3.13	-90.00	0.2047	0.00	-0.05	0.10	-0.01	0.6927	0.6416	0.0015	0.0002	0.0000	0.0073	0.2275	0.6561
	8	0.0283	0.0013	0.0030	0.2593	0.5715	0.0038	0.0000	0.0005								
	9	0.0262	0.0013	0.0030	0.2593	0.5715	0.0038	0.0000	0.0005								
709	4	0.899	-3.13	-78.80	0.2048	0.00	-0.06	0.10	-0.02	0.1872	0.6411	0.0016	0.0000	0.0003	0.12236	1.0883	1.5027
	8	0.0278	0.0011	0.0035	0.2654	0.5844	-0.0008	-0.0002	-0.0002								
	9	0.0256	0.0013	0.0029	0.2654	0.5844	-0.0008	-0.0002	-0.0002								
709	5	0.900	-3.13	-67.50	0.2139	0.00	-0.07	0.10	-0.03	0.1536	0.6331	0.0069	0.0002	0.0010	0.1536	0.7521	0.8083
	8	0.0264	0.0011	0.0033	0.3033	0.6331	-0.0054	-0.0004	-0.0008								
	9	0.0226	0.0013	0.0028	0.3033	0.6331	-0.0054	-0.0004	-0.0008								
709	6	0.901	-3.14	-56.30	0.2165	0.00	-0.08	0.09	-0.04	0.1133	0.6347	0.0134	0.0003	0.0019	0.1133	0.7120	0.7371
	8	0.0238	0.0010	0.0030	0.2868	0.5745	-0.0102	-0.0007	-0.0015								
	9	0.0209	0.0012	0.0024	0.2868	0.5745	-0.0102	-0.0007	-0.0015								
709	7	0.900	-3.14	-45.00	0.2317	0.00	-0.09	0.09	-0.05	0.1306	0.6558	0.0185	0.0004	0.0025	0.1306	0.6761	0.6833
	8	0.0201	0.0009	0.0026	0.2909	0.5413	-0.0151	-0.0009	-0.0020								
	9	0.0173	0.0010	0.0018	0.2909	0.5413	-0.0151	-0.0009	-0.0020								
709	8	0.899	-3.14	-33.80	0.1519	0.01	-0.09	0.08	-0.05	0.1515	0.5882	0.0225	0.0006	0.0030	0.1515	0.6671	0.6720
	8	0.0174	0.0005	0.0020	0.3737	0.6078	-0.0191	-0.0011	-0.0025								
	9	0.0116	0.0008	0.0014	0.3737	0.6078	-0.0191	-0.0011	-0.0025								
709	9	0.898	-3.14	-22.50	0.1783	0.02	-0.10	0.07	-0.06	0.1755	0.6627	0.0252	0.0008	0.0033	0.1755	0.6546	0.6665
	8	0.0118	0.0004	0.0015	0.4984	0.5980	-0.0220	-0.0012	-0.0029								
	9	0.0066	0.0006	0.0007	0.4984	0.5980	-0.0220	-0.0012	-0.0029								
709	10	0.899	-3.14	-11.30	0.2224	0.01	-0.11	0.06	-0.07	0.1907	0.8672	0.0269	0.0010	0.0035	0.1907	0.6539	0.6595
	8	0.0058	0.0002	0.0010	0.9035	0.3776	-0.0239	-0.0013	-0.0031								
	9	0.0021	0.0003	0.0001	0.9035	0.3776	-0.0239	-0.0013	-0.0031								
709	11	0.900	-3.14	-0.00	-0.2449	-0.03	-0.10	0.05	-0.06	0.1944	1.3643	0.0278	0.0010	0.0035	0.1944	0.6423	0.6548
	8	0.0016	-0.0000	0.0004	-0.4626	1.5123	-0.0253	-0.0013	-0.0033								
	9	-0.0010	0.0001	-0.0003	-0.4626	1.5123	-0.0253	-0.0013	-0.0033								

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd																			
709 12	7 8 9	0.898 -0.0027 -0.0065	-3.14 -0.0003 -0.0000	11.29 -0.0003 -0.0006	0.6167 0.0158	-0.03 0.5459 0.4857	-0.10 -0.0266 -0.0250	0.04 -0.0011 -0.0012	-0.06 -0.0035 -0.0032	0.2118 0.2576	0.6601 0.6527									
709 13	7 8 9	0.900 -0.0083 -0.0108	-3.14 -0.0005 -0.0003	22.50 -0.0009 -0.0013	0.3546 0.1402	-0.05 0.5676 0.6125	-0.10 -0.0249 -0.0236	0.03 -0.0010 -0.0011	-0.05 -0.0032 -0.0031	0.2183 0.2531	0.6596 0.6615									
709 14	7 8 9	0.898 -0.0137 -0.0173	-3.14 -0.0008 -0.0004	33.79 -0.0015 -0.0020	0.3098 0.1296	-0.06 0.5660 0.6053	-0.10 -0.0225 -0.0211	0.03 -0.0009 -0.0010	-0.05 -0.0029 -0.0028	0.2159 0.2479	0.6612 0.6719									
709 15	7 8 9	0.898 -0.0165 -0.0212	-3.14 -0.0011 -0.0007	44.99 -0.0021 -0.0028	0.2979 0.1773	-0.06 0.5824 0.6724	-0.09 -0.0189 -0.0175	0.02 -0.0008 -0.0008	-0.05 -0.0025 -0.0024	0.2132 0.2498	0.6672 0.6908									
709 16	7 8 9	0.900 -0.0217 -0.0262	-3.14 -0.0013 -0.0008	56.29 -0.0026 -0.0031	0.3213 0.1697	-0.08 0.6139 0.6052	-0.08 -0.0148 -0.0129	0.01 -0.0006 -0.0006	-0.05 -0.0020 -0.0018	0.2042 0.2670	0.6970 0.7321									
709 17	7 8 9	0.899 -0.0261 -0.0275	-3.14 -0.0013 -0.0012	67.49 -0.0029 -0.0036	0.2625 0.2182	-0.08 0.5676 0.6684	-0.08 -0.0103 -0.0079	0.00 -0.0003 -0.0004	-0.04 -0.0015 -0.0012	0.1907 0.3146	0.7270 0.7930									
709 18	7 8 9	0.900 -0.0276 -0.0315	-3.14 -0.0014 -0.0011	78.79 -0.0032 -0.0036	0.2704 0.1893	-0.08 0.5841 0.5778	-0.07 -0.0049 -0.0034	0.00 -0.0002 -0.0002	-0.03 -0.0008 -0.0007	0.2238 0.4278	0.8407 1.0051									
709 19	7 8 9	0.899 -0.0280 -0.0319	-3.14 -0.0016 -0.0012	89.99 -0.0034 -0.0037	0.2867 0.1994	-0.09 0.6069 0.5811	-0.06 -0.0002 0.0005	0.00 -0.0000 -0.0000	-0.03 -0.0002 -0.0001	-0.1551 -0.5909	3.7671 -1.5960									
710 1	7 8 9	0.900 0.0001 -0.0012	-0.04 -0.0003 0.0002	89.99 0.0001 -0.0001	-0.7764 -0.8297	-0.04 3.7057 0.7275	-0.06 -0.0018 0.0000	0.06 -0.0000 -0.0001	-0.02 -0.0004 -0.0001	0.2231 -38.6237	1.1944 -39.7876									
710 2	7 8 9	0.899 -0.0003 -0.0015	-0.05 -0.0002 0.0002	78.79 0.0001 -0.0001	4.5314 -0.7337	-0.04 -1.8563 0.4451	-0.06 -0.0019 0.0002	0.05 -0.0000 -0.0001	-0.03 -0.0003 -0.0000	0.1473 -2.4838	1.0269 -2.0881									
710 3	7 8 9	0.899 -0.0007 -0.0013	-0.05 -0.0001 0.0002	67.49 0.0001 -0.0001	1.2264 -0.7940	-0.03 -1.2045 0.5632	-0.07 -0.0013 0.0004	0.05 -0.0000 -0.0000	-0.02 -0.0003 -0.0000	0.1018 -0.9474	1.2259 -0.4315									

See Key

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																				
710	15	0.898	-0.005	-67.50	1.3817	-0.04	-0.05	0.05	0.002	-0.02	0.5123	0.4081	-0.009	-0.0002	-0.0001	1.3817	-0.04	-0.05	0.002	-0.02	0.5123	0.4081
	8	-0.0019	0.0000	-0.0003	-0.1250	0.9356	0.0041	0.0000	0.0000	0.0005	0.0428	0.6479	-0.0019	0.0000	-0.0003	-0.1250	0.9356	0.0041	0.0000	0.0005	0.0428	0.6479
710	16	0.898	-0.006	-78.80	1.2017	-0.04	-0.06	0.05	0.001	-0.02	0.4191	0.3761	-0.0012	-0.0002	-0.0001	1.2017	-0.04	-0.06	0.001	-0.02	0.4191	0.3761
	8	-0.0017	0.0000	-0.0003	-0.1374	1.0946	0.0048	0.0000	0.0000	0.0005	0.0386	0.6099	-0.0017	0.0000	-0.0003	-0.1374	1.0946	0.0048	0.0000	0.0005	0.0386	0.6099
710	17	0.898	-0.005	-90.00	1.6670	-0.04	-0.06	0.05	0.001	-0.02	0.3478	0.2535	-0.0010	-0.0003	-0.0002	1.6670	-0.04	-0.06	0.001	-0.02	0.3478	0.2535
	8	-0.0026	0.0001	-0.0002	-0.2572	0.4441	0.0050	0.0000	0.0000	0.0005	0.0285	0.5653	-0.0026	0.0001	-0.0002	-0.2572	0.4441	0.0050	0.0000	0.0005	0.0285	0.5653
711	1	0.898	3.07	-90.00	0.3229	-0.09	-0.06	0.00	0.00	-0.01	0.4276	0.3394	-0.0299	-0.0019	-0.0040	0.3229	-0.09	-0.06	0.00	-0.01	0.4276	0.3394
	8	-0.0307	-0.0014	-0.0039	0.2437	0.6497	0.0037	0.0001	0.0001	0.0003	0.1415	0.4239	-0.0307	-0.0014	-0.0039	0.2437	0.6497	0.0037	0.0001	0.0003	0.1415	0.4239
711	2	0.899	3.06	-78.80	0.2949	-0.09	-0.04	0.00	0.00	-0.01	0.3286	0.5815	-0.0289	-0.0017	-0.0037	0.2949	-0.09	-0.04	0.00	-0.01	0.3286	0.5815
	8	-0.0289	-0.0014	-0.0037	0.2467	0.6429	0.0083	0.0002	0.0002	0.0010	0.1778	0.6126	-0.0289	-0.0017	-0.0037	0.2467	0.6429	0.0083	0.0002	0.0010	0.1778	0.6126
711	3	0.898	3.05	-67.50	0.3143	-0.09	-0.04	0.00	0.00	-0.00	0.2689	0.6573	-0.0277	-0.0016	-0.0033	0.3143	-0.09	-0.04	0.00	-0.00	0.2689	0.6573
	8	-0.0277	-0.0016	-0.0033	0.2297	0.6090	0.0154	0.0003	0.0003	0.0019	0.1282	0.6212	-0.0277	-0.0016	-0.0033	0.2297	0.6090	0.0154	0.0003	0.0019	0.1282	0.6212
711	4	0.899	3.05	-56.30	0.2661	-0.08	-0.02	0.01	0.01	-0.00	0.2625	0.6551	-0.0233	-0.0012	-0.0030	0.2661	-0.08	-0.02	0.01	-0.00	0.2625	0.6551
	8	-0.0248	-0.0011	-0.0030	0.2231	0.6055	0.0213	0.0004	0.0004	0.0027	0.1104	0.6427	-0.0248	-0.0011	-0.0030	0.2231	0.6055	0.0213	0.0004	0.0004	0.1104	0.6427
711	5	0.897	3.05	-45.00	0.2519	-0.07	-0.02	0.02	0.02	0.00	0.2539	0.6466	-0.0189	-0.0009	-0.0022	0.2519	-0.07	-0.02	0.02	0.00	0.2539	0.6466
	8	-0.0186	-0.0010	-0.0028	0.2949	0.7661	0.0257	0.0006	0.0006	0.0033	0.1270	0.6411	-0.0186	-0.0009	-0.0022	0.2949	0.7661	0.0257	0.0006	0.0006	0.1270	0.6411
711	6	0.898	3.05	-33.80	0.3010	-0.06	-0.01	0.03	0.03	0.00	0.2450	0.6398	-0.0129	-0.0007	-0.0020	0.3010	-0.06	-0.01	0.03	0.00	0.2450	0.6398
	8	-0.0159	-0.0007	-0.0020	0.2229	0.6543	0.0287	0.0008	0.0008	0.0037	0.1460	0.6554	-0.0159	-0.0007	-0.0020	0.2229	0.6543	0.0287	0.0008	0.0008	0.1460	0.6554
711	7	0.897	3.06	-22.50	0.3632	-0.05	-0.01	0.03	0.03	0.00	0.2430	0.6456	-0.0073	-0.0005	-0.0008	0.3632	-0.05	-0.01	0.03	0.00	0.2430	0.6456
	8	-0.0114	-0.0003	-0.0013	0.1601	0.6060	0.0303	0.0010	0.0010	0.0039	0.1678	0.6479	-0.0114	-0.0003	-0.0008	0.1601	0.6060	0.0303	0.0010	0.0010	0.1678	0.6479
711	8	0.898	3.05	-11.30	0.7424	-0.03	-0.01	0.05	0.05	0.01	0.2331	0.6456	-0.0019	-0.0002	-0.0006	0.7424	-0.03	-0.01	0.05	0.01	0.2331	0.6456
	8	-0.0064	-0.0000	-0.0006	0.0475	0.5221	0.0308	0.0010	0.0010	0.0039	0.1727	0.6377	-0.0064	-0.0000	-0.0006	0.0475	0.5221	0.0308	0.0010	0.0010	0.1727	0.6377

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																
711	9	7	0.897	3.06	-0.00	-0.1934	-0.01	-0.01	0.06	0.01	0.2253	0.6494	0.0034	-0.0001	0.0004	0.0014	0.0041	0.2253	0.6494
711	10	7	0.897	3.06	11.29	0.0224	-0.00	-0.01	0.06	0.01	0.2136	0.6486	0.0090	0.0000	0.0011	0.0013	0.0039	0.2136	0.6486
711	11	7	0.898	3.06	22.49	0.0810	-0.00	-0.02	0.08	0.01	0.2023	0.6498	0.0153	0.0002	0.0011	0.0011	0.0036	0.2023	0.6498
711	12	7	0.898	3.06	33.79	0.1451	0.01	-0.03	0.09	0.00	0.1951	0.6517	0.0199	0.0005	0.0009	0.0008	0.0031	0.1951	0.6517
711	13	7	0.897	3.06	44.99	0.1960	0.01	-0.04	0.10	0.00	0.1663	0.6556	0.0234	0.0009	0.0014	0.0017	0.0024	0.1663	0.6556
711	14	7	0.896	3.06	56.29	0.1862	0.02	-0.04	0.10	0.01	0.1653	0.6610	0.0270	0.0010	0.0016	0.0015	0.0018	0.1653	0.6610
711	15	7	0.898	3.06	67.49	0.2005	0.02	-0.05	0.11	0.01	0.4529	0.5886	0.0284	0.0011	0.0016	0.0005	0.0012	0.4529	0.5886
711	16	7	0.899	3.05	78.79	0.2135	0.02	-0.06	0.11	0.02	0.4792	0.3951	0.0290	0.0012	0.0016	0.0002	0.0005	0.4792	0.3951
711	17	7	0.899	3.06	89.99	0.2060	0.02	-0.06	0.11	0.03	0.5470	1.2031	0.0285	0.0011	0.0016	0.0001	0.0002	0.5470	1.2031
712	1	7	0.899	6.17	89.99	0.2145	0.08	-0.06	0.16	0.02	-5.0269	-5.6542	0.0596	0.0025	0.0028	0.0001	0.0003	-5.0269	-5.6542
712	2	7	0.898	6.16	78.79	0.2047	0.08	-0.04	0.16	0.02	0.3025	0.6280	0.0601	0.0024	0.0029	0.0004	0.0009	0.3025	0.6280

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																											
712	3	0.898	6.17	67.49	0.1769	0.07	-0.03	0.15	0.00	0.1270	0.6525	0.0602	0.0021	0.0076	0.1769	0.6368	0.0217	0.0005	0.0028	0.2146	0.6343	0.0515	0.0029	0.0063	0.2829	0.6124	0.0191	0.0008	0.0024
712	4	0.898	6.16	56.29	0.1632	0.06	-0.01	0.15	0.00	0.1367	0.6460	0.0550	0.0017	0.0070	0.1632	0.6433	0.0333	0.0009	0.0043	0.2275	0.6245	0.0476	0.0027	0.0057	0.2863	0.6066	0.0296	0.0013	0.0037
712	5	0.899	6.15	44.99	0.1570	0.04	-0.00	0.13	0.02	0.1512	0.6520	0.0464	0.0014	0.0060	0.1570	0.6514	0.0438	0.0013	0.0057	0.2295	0.6291	0.0416	0.0023	0.0049	0.2824	0.5907	0.0396	0.0018	0.0049
712	6	0.896	6.15	33.79	0.1415	0.03	0.01	0.12	0.04	0.1686	0.6375	0.0363	0.0019	0.0040	0.1415	0.6587	0.0515	0.0017	0.0065	0.2234	0.6261	0.0328	0.0019	0.0040	0.3033	0.6138	0.0477	0.0021	0.0059
712	7	0.898	6.17	22.49	0.1157	0.01	0.02	0.10	0.05	0.1885	0.6204	0.0249	0.0005	0.0032	0.1157	0.6470	0.0559	0.0021	0.0069	0.2139	0.6116	0.0226	0.0014	0.0026	0.3264	0.5947	0.0540	0.0023	0.0066
712	8	0.900	6.16	11.29	0.0766	0.00	0.02	0.07	0.05	0.2082	0.6119	0.0127	0.0001	0.0017	0.0766	0.6705	0.0577	0.0024	0.0070	0.2081	0.6104	0.0103	0.0008	0.0012	0.4215	0.6066	0.0577	0.0024	0.0070
712	9	0.898	6.17	-0.00	-0.2453	-0.00	0.03	0.05	0.05	0.2192	0.6067	-0.0000	0.0000	-0.0000	-0.2453	-0.4772	0.0587	0.0025	0.0073	0.1927	0.6042	0.0004	0.0002	0.0000	-2.7199	-0.3715	0.0606	0.0023	0.0073
712	10	0.898	6.17	-11.30	0.2116	0.04	0.03	0.03	0.05	0.2274	0.6075	-0.0101	-0.0004	-0.0012	0.2116	0.6049	0.0580	0.0026	0.0070	0.1817	0.6075	0.0109	-0.0003	-0.0011	0.1581	0.5430	0.0608	0.0022	0.0073
712	11	0.897	6.16	-22.50	0.1789	0.06	0.03	0.00	0.04	0.2361	0.6090	-0.0222	-0.0007	-0.0025	0.1789	0.6022	0.0555	0.0019	0.0073	0.1668	0.6199	0.0213	-0.0010	-0.0025	0.2420	0.6077	0.0595	0.0019	0.0073
712	12	0.898	6.16	-33.80	0.1914	0.09	0.02	0.00	0.00	0.2449	0.6149	-0.0335	-0.0012	-0.0042	0.1914	0.6328	0.0506	0.0024	0.0062	0.1488	0.6287	0.0299	-0.0018	-0.0041	0.3019	0.6849	0.0551	0.0016	0.0069
712	13	0.897	6.16	-45.00	0.2034	0.10	0.01	0.03	0.03	0.2508	0.6272	-0.0432	-0.0017	-0.0055	0.2034	0.6441	0.0443	0.0022	0.0060	0.1358	0.6354	0.0406	-0.0022	-0.0051	0.2711	0.6375	0.0475	0.0012	0.0060

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key											
712	14	7	0.897	6.15	-56.30	0.2192	-0.12	-0.00	-0.03	0.01	0.2637	0.6366	
		8	-0.0498	-0.0021	-0.0061	0.2546	0.6142	0.0348	0.0018	0.0044	0.1104	0.6300	
		9	-0.0495	-0.0025	-0.0060		0.6127	0.0379	0.0008	0.0047			
712	15	7	0.895	6.16	-67.50	0.2403	-0.13	-0.02	-0.04	-0.00	0.2613	0.6328	
		8	-0.0535	-0.0025	-0.0064	0.2484	0.6050	0.0244	0.0012	0.0030	0.0808	0.6221	
		9	-0.0555	-0.0027	-0.0068		0.6206	0.0264	0.0004	0.0032			
712	16	7	0.896	6.16	-78.80	0.2504	-0.10	-0.04	-0.05	-0.01	0.2613	0.6274	
		8	-0.0559	-0.0028	-0.0067	0.2510	0.6000	0.0136	0.0007	0.0017	0.1510	0.5835	
		9	-0.0608	-0.0028	-0.0073		0.6052	0.0120	0.0003	0.0014			
712	17	7	0.898	6.17	-90.00	0.2544	-0.13	-0.05	-0.05	-0.03	0.2789	0.5713	
		8	-0.0570	-0.0029	-0.0068	0.2182	0.5967	0.0028	0.0001	0.0002	0.1606	-0.1315	
		9	-0.0630	-0.0027	-0.0075		0.5986	0.0019	0.0000	-0.0000			
713	1	7	0.899	9.31	-90.00	0.2125	-0.16	-0.06	-0.07	-0.02	0.2125	0.5986	
		8	-0.0800	-0.0034	-0.0090	0.1945	0.5629	0.0023	0.0000	0.0001	-0.1448	0.5230	
		9	-0.0831	-0.0032	-0.0094		0.5680	0.0030	-0.0000	0.0001			
713	2	7	0.898	9.30	-78.79	0.2057	-0.16	-0.03	-0.07	-0.00	0.2709	0.6486	
		8	-0.0802	-0.0033	-0.0091	0.2061	0.5688	0.0168	0.0009	0.0021	0.0948	0.5950	
		9	-0.0807	-0.0033	-0.0091		0.5690	0.0177	0.0003	0.0021			
713	3	7	0.898	9.30	-78.79	0.2064	-0.16	-0.03	-0.07	-0.00	0.2676	0.6410	
		8	-0.0801	-0.0033	-0.0091	0.2029	0.5681	0.0176	0.0009	0.0022	0.0954	0.5880	
		9	-0.0810	-0.0032	-0.0091		0.5640	0.0184	0.0003	0.0021			
713	4	7	0.898	9.29	-67.49	0.2085	-0.15	-0.00	-0.07	0.01	0.2712	0.6348	
		8	-0.0773	-0.0032	-0.0089	0.2222	0.5793	0.0333	0.0018	0.0043	0.1265	0.6325	
		9	-0.0757	-0.0033	-0.0085		0.5635	0.0352	0.0008	0.0043			
713	5	7	0.900	9.28	-56.29	0.1929	-0.14	0.02	-0.06	0.03	0.2619	0.6080	
		8	-0.0723	-0.0027	-0.0085	0.2390	0.5891	0.0481	0.0025	0.0058	0.1265	0.6325	
		9	-0.0684	-0.0032	-0.0075		0.5544	0.0528	0.0013	0.0066			
713	6	7	0.899	9.28	-45.00	0.1933	-0.12	0.03	-0.05	0.04	0.2463	0.5932	
		8	-0.0625	-0.0024	-0.0075	0.2568	0.5999	0.0600	0.0029	0.0071	0.1308	0.6198	
		9	-0.0594	-0.0030	-0.0065		0.5531	0.0679	0.0017	0.0084			
713	7	7	0.898	9.29	-33.80	0.1844	-0.10	0.04	-0.03	0.06	0.2340	0.5778	
		8	-0.0504	-0.0018	-0.0062	0.2976	0.6177	0.0677	0.0031	0.0078	0.1460	0.5969	
		9	-0.0459	-0.0027	-0.0059		0.6492	0.0764	0.0022	0.0091			

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																			
713	8	0.899	9.30	-22.50	0.1911	-0.09	0.05	-0.00	0.07	0.2137	0.5748	0.0331	-0.0012	-0.0039	0.1911	0.5994	0.0744	-0.0031	0.0085	0.1646	0.5879
	8	-0.0316	-0.0018	-0.0040	0.2863	0.6440	0.0810	0.0026	0.0095												
713	9	0.898	9.30	-11.30	0.2401	-0.05	0.05	0.02	0.08	0.1995	0.5740	0.0151	-0.0007	-0.0016	0.2401	0.5584	0.0787	0.0031	0.0090	0.1777	0.5765
	9	-0.0152	-0.0008	-0.0020	0.2740	0.6824	0.0827	0.0029	0.0095												
713	10	0.898	9.31	-0.00	-2.1031	-0.02	0.05	0.06	0.08	0.1916	0.5762	0.0000	-0.0000	0.0001	-2.1031	14.2188	0.0807	0.0030	0.0093	0.1843	0.5687
	8	-0.0010	0.0002	-0.0000	-1.0482	0.1194	0.0824	0.0030	0.0093												
713	11	0.897	9.29	11.30	0.1679	0.00	0.05	0.09	0.08	0.1905	0.5796	0.0162	0.0005	0.0021	0.1679	0.6474	0.0804	0.0030	0.0093	0.1961	0.5705
	8	0.0139	0.0011	0.0018	0.4171	0.6816	0.0797	0.0031	0.0090												
713	12	0.899	9.30	22.50	0.1434	0.02	0.05	0.12	0.07	0.1810	0.5884	0.0345	0.0009	0.0044	0.1434	0.6429	0.0779	0.0028	0.0091	0.2085	0.5710
	9	0.0310	0.0020	0.0040	0.3352	0.6490	0.0750	0.0031	0.0085												
713	13	0.896	9.28	33.79	0.1433	0.05	0.04	0.15	0.07	0.1635	0.5981	0.0529	0.0015	0.0068	0.1433	0.6507	0.0737	0.0024	0.0088	0.2218	0.5782
	8	0.0448	0.0028	0.0055	0.3192	0.6234	0.0680	0.0030	0.0078												
713	14	0.896	9.29	44.99	0.1429	0.07	0.02	0.17	0.06	0.1494	0.6164	0.0679	0.0019	0.0056	0.1429	0.6444	0.0844	0.0019	0.0068	0.2374	0.5881
	9	0.0571	0.0033	0.0066	0.2896	0.5844	0.0582	0.0027	0.0068												
713	15	0.896	9.28	56.29	0.1608	0.09	0.00	0.18	0.04	0.1465	0.6288	0.0758	0.0024	0.0091	0.1608	0.6066	0.0499	0.0014	0.0052	0.2433	0.6164
	9	0.0663	0.0035	0.0077	0.2682	0.5808	0.0463	0.0022	0.0057												
713	16	0.897	9.29	67.49	0.1761	0.10	-0.02	0.19	0.01	0.1491	0.6164	0.0793	0.0027	0.0083	0.1761	0.5979	0.0319	0.0009	0.0037	0.2427	0.6209
	9	0.0724	0.0035	0.0083	0.2467	0.5766	0.0305	0.0014	0.0037												
713	17	0.899	9.29	78.79	0.1901	0.10	-0.04	0.19	-0.00	0.1495	0.6024	0.0801	0.0030	0.0094	0.1901	0.5880	0.0143	0.0004	0.0017	0.2022	0.6013
	9	0.0768	0.0035	0.0087	0.2289	0.5681	0.0140	0.0005	0.0016												
713	18	0.897	9.29	89.99	0.1993	0.10	-0.06	0.19	-0.02	0.3993	2.489	0.0796	0.0031	0.0089	0.1993	0.5860	0.0006	0.0000	0.0002	0.3993	2.489
	9	0.0788	0.0034	0.0089	0.2184	0.5661	0.0004	-0.0002	-0.0002												

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key													
714	1	7 8 9	0.897 0.0974 0.0951	12.41 0.0030 0.0033	90.00 0.0110 0.0106	0.1584 0.1763	0.5648 0.5617	0.12 -0.0009 0.0006	-0.06 -0.0006	0.20 -0.0002 -0.0001	-0.02 -0.0003 -0.0001	0.1249 -1.6865	1.5983 -1.5049			
714	2	7 8 9	0.898 0.1007 0.0926	12.42 0.0031 0.0032	78.78 0.0115 0.0102	0.1574 0.1771	0.5717 0.5544	0.12 -0.03 0.0195 0.0197	0.20 0.0005 0.0008	0.00 0.0022 0.0024	0.1494 0.2244	0.5869 0.6106				
714	3	7 8 9	0.896 0.0996 0.0857	12.41 0.0030 0.0034	67.48 0.0114 0.0097	0.1544 0.2028	0.5762 0.5686	0.12 -0.00 0.0430 0.0416	0.20 0.0012 0.0020	0.03 0.0053 0.0051	0.1424 0.2490	0.6222 0.6234				
714	4	7 8 9	0.897 0.0951 0.0795	12.41 0.0027 0.0036	56.29 0.0112 0.0087	0.1464 0.2287	0.5910 0.5475	0.11 0.02 0.0663 0.0580	0.20 0.0017 0.0029	0.06 0.0063 0.0067	0.1316 0.2527	0.6318 0.5813				
714	5	7 8 9	0.899 0.0857 0.0718	12.41 0.0023 0.0036	44.99 0.0106 0.0078	0.1384 0.2544	0.6210 0.5462	0.09 0.04 0.0840 0.0721	0.19 0.0022 0.0032	0.08 0.0103 0.0080	0.1349 0.2257	0.6171 0.5597				
714	6	7 8 9	0.897 0.0696 0.0560	12.42 0.0017 0.0034	33.80 0.0091 0.0065	0.1253 0.3089	0.6540 0.5798	0.07 0.06 0.0943 0.0828	0.17 0.0027 0.0032	0.08 0.0112 0.0090	0.1436 0.1976	0.5947 0.5469				
714	7	7 8 9	0.898 0.0447 0.0402	10.39 0.0012 0.0025	22.51 0.0058 0.0048	0.1385 0.3179	0.604 0.6518 0.6020	0.04 0.07 0.0987 0.0902	0.14 0.0030 0.0031	0.08 0.0114 0.0099	0.1546 0.1733	0.5792 0.5503				
714	8	7 8 9	0.898 0.0215 0.0182	12.42 0.0005 0.0014	11.31 0.0026 0.0023	0.1264 0.3954	0.6145 0.6577	0.07 0.001 0.0957	0.09 0.0031 0.0029	0.09 0.0114 0.0105	0.1557 0.1557	0.5700 0.5512				
714	9	7 8 9	0.899 0.0005 -0.0007	12.42 -0.0001 0.0001	0.00 0.0002 -0.0002	-1.3031 -0.8081	-0.02 1.9509 1.6665	0.07 0.0973 0.0994	0.04 0.0030 0.0029	0.08 0.0109 0.0110	0.1549 0.1500	0.5625 0.5534				
714	10	7 8 9	0.897 -0.0213 -0.0216	12.42 -0.0009 -0.0011	-11.31 -0.0024 -0.0027	0.2151 0.2712	-0.06 0.5770 0.6405	0.06 0.0940 0.1025	0.00 0.0029 0.0031	0.09 0.0105 0.0115	0.1588 0.1514	0.5611 0.5609				
714	11	7 8 9	0.899 -0.0441 -0.0434	12.42 -0.0016 -0.0023	-22.51 -0.0055 -0.0052	0.1835 0.2689	-0.10 0.6306 0.5990	0.07 0.0883 0.1008	-0.03 0.0030 0.0029	0.09 0.0098 0.0115	0.1748 0.1483	0.5584 0.5713				

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key														
714	12	0.900	12.42	-33.80	0.1585	-0.13	0.06	-0.07	0.08	0.1994	0.5556					
	8	-0.0666	-0.0021	-0.0082	0.2740	0.6155	0.0810	0.0032	0.0090	0.1357	0.5825					
	9	-0.0584	-0.0032	-0.0065		0.5613	0.0959	0.0026	0.0111							
714	13	0.898	12.41	-45.00	0.1603	-0.16	0.05	-0.07	0.06	0.2259	0.5639					
	8	-0.0833	-0.0026	-0.0101	0.2309	0.6063	0.0734	0.0033	0.0082	0.1172	0.6080					
	9	-0.0734	-0.0033	-0.0081		0.5516	0.0879	0.0020	0.0106							
714	14	0.897	12.42	-56.29	0.1678	-0.17	0.04	-0.09	0.04	0.2650	0.5820					
	8	-0.0944	-0.0031	-0.0111	0.2020	0.5909	0.0592	0.0031	0.0068	0.1074	0.6310					
	9	-0.0817	-0.0033	-0.0091		0.5602	0.0698	0.0015	0.0088							
714	15	0.899	12.42	-67.49	0.1695	-0.18	0.01	-0.09	0.02	0.2708	0.6143					
	8	-0.0990	-0.0033	-0.0112	0.1673	0.5699	0.0423	0.0022	0.0052	0.1129	0.6129					
	9	-0.0899	-0.0030	-0.0098		0.5488	0.0463	0.0010	0.0056							
714	16	0.897	12.43	-78.79	0.1654	-0.18	-0.02	-0.09	-0.00	0.2679	0.6220					
	8	-0.1002	-0.0033	-0.0111	0.1544	0.5554	0.0213	0.0011	0.0026	0.0825	0.5764					
	9	-0.0947	-0.0029	-0.0106		0.5597	0.0234	0.0003	0.0027							
714	17	0.897	12.42	-90.00	0.1610	-0.18	-0.06	-0.10	-0.03	0.1909	0.0753					
	8	-0.0985	-0.0031	-0.0108	0.1533	0.5520	0.0015	0.0000	0.0000	-0.2642	0.4029					
	9	-0.0981	-0.0030	-0.0111		0.5676	0.0036	-0.0001	0.0002							
715	1	0.898	12.43	-90.00	0.1625	-0.17	-0.05	-0.09	3.06	0.2434	0.2482					
	8	-0.0996	-0.0032	-0.0110	0.1527	0.5554	0.0018	0.0000	0.0000	0.0571	0.5897					
	9	-0.0971	-0.0029	-0.0109		0.5631	0.0222	0.0002	0.0026							
715	2	0.899	12.43	-78.79	0.1732	-0.19	-0.02	-0.09	3.08	0.2743	0.6341					
	8	-0.1009	-0.0034	-0.0114	0.1522	0.5674	0.0218	0.0011	0.0027	0.0903	0.6206					
	9	-0.0938	-0.0028	-0.0103		0.5489	0.0453	0.0008	0.0056							
715	3	0.898	12.43	-67.49	0.1672	-0.18	0.01	-0.09	3.10	0.2726	0.6177					
	8	-0.1007	-0.0033	-0.0115	0.1702	0.5747	0.0427	0.0023	0.0052	0.0939	0.6331					
	9	-0.0886	-0.0030	-0.0097		0.5501	0.0687	0.0012	0.0087							
715	4	0.897	12.43	-56.29	0.1597	-0.17	0.04	-0.08	3.13	0.2644	0.5835					
	8	-0.0964	-0.0030	-0.0114	0.2002	0.5956	0.0595	0.0031	0.0069	0.1046	0.6180					
	9	-0.0815	-0.0032	-0.0089		0.5465	0.0891	0.0018	0.0110							
715	5	0.896	12.42	-44.99	0.1504	-0.15	0.05	-0.07	3.14	0.2243	0.5644					
	8	-0.0847	-0.0025	-0.0103	0.2415	0.6077	0.0740	0.0033	0.0083	0.1239	0.5840					
	9	-0.0716	-0.0034	-0.0080		0.5615	0.1004	0.0024	0.0117							

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

← See Key →

Run Pt.	Pt.	Cd																					
715	6	7 8 9	0.899	12.43	-33.80	0.1497	-0.13	0.06	-0.06	3.15	0.1990	0.5567											
			-0.0671	-0.0020	-0.0082	0.6175	0.0616	0.0032	0.0123	0.0029													
			-0.0572	-0.0031	-0.0063	0.5522	0.1080	0.0029															
715	7	7 8 9	0.899	12.43	-22.51	0.1885	-0.09	0.06	-0.03	3.16	0.1732	0.5572											
			-0.0427	-0.0016	-0.0053	0.6285	0.0690	0.0030	0.0122	0.0029													
			-0.0420	-0.0023	-0.0050	0.6015	0.1095	0.0029															
715	8	7 8 9	0.898	12.43	-11.31	0.2287	-0.06	0.07	0.00	3.16	0.1606	0.5628											
			-0.0200	-0.0009	-0.0022	0.5651	0.0942	0.0030	0.0122	0.0027													
			-0.0202	-0.0011	-0.0026	0.6625	0.1096	0.0027															
715	9	7 8 9	0.898	12.43	0.00	-0.2204	-0.01	0.06	0.05	3.15	0.1561	0.5658											
			-0.0013	-0.0000	0.0003	1.3680	0.0982	0.0030	0.0111	0.0027													
			-0.0003	-0.0002	-0.0000	0.6905	0.1062	0.0027															
715	10	7 8 9	0.897	12.43	11.31	0.1428	0.01	0.07	0.09	3.15	0.1547	0.5718											
			0.0222	0.0006	0.0027	0.268	0.1010	0.0031	0.0113	0.0027													
			0.0201	0.0014	0.0025	0.6329	0.1021	0.0027															
715	11	7 8 9	0.900	12.42	22.51	0.1427	0.04	0.07	0.13	3.15	0.1535	0.5796											
			0.0450	0.0012	0.0058	0.6500	0.0989	0.0030	0.0114	0.0027													
			0.0404	0.0026	0.0050	0.6187	0.0970	0.0027															
715	12	7 8 9	0.899	12.42	33.80	0.1307	0.07	0.06	0.17	3.15	0.1418	0.5941											
			0.0695	0.0018	0.0090	0.6528	0.0946	0.0026	0.0112	0.0029													
			0.0573	0.0034	0.0065	0.5693	0.0901	0.0029															
715	13	7 8 9	0.898	12.42	44.99	0.1365	0.10	0.04	0.19	3.15	0.1342	0.6111											
			0.0853	0.0023	0.0105	0.6165	0.0845	0.0022	0.0103	0.0031													
			0.0716	0.0037	0.0079	0.5569	0.0806	0.0031															
715	14	7 8 9	0.899	12.41	56.29	0.1524	0.11	0.02	0.19	3.14	0.1298	0.6271											
			0.0939	0.0028	0.0112	0.5977	0.0663	0.0017	0.0077	0.0032													
			0.0809	0.0035	0.0087	0.5381	0.0693	0.0032															
715	15	7 8 9	0.899	12.41	67.48	0.1573	0.12	-0.00	0.20	3.12	0.1407	0.6155											
			0.0989	0.0031	0.0114	0.5775	0.0434	0.0012	0.0053	0.0028													
			0.0877	0.0033	0.0096	0.5511	0.0543	0.0028															
715	16	7 8 9	0.898	12.42	78.78	0.1643	0.12	-0.04	0.20	3.09	0.1430	0.5693											
			0.0989	0.0032	0.0114	0.5767	0.0202	0.0005	0.0023	0.0019													
			0.0932	0.0032	0.0102	0.5507	0.0368	0.0019															

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																					
715	17	0.897	12.42	89.99	0.1568	0.5629	0.12	-0.06	-0.0005	0.20	3.05	0.3805	2.7059	0.0978	0.0030	0.0110	0.012	-0.0005	-0.0000	0.0003	0.2326	0.6376	
	8	0.0957	0.0033	0.0108	0.1770	0.5667	0.0149	0.0006	0.0019	0.0019	0.0019	0.0019	0.0019	0.0019	0.0019	0.0019	0.0019	0.0019	0.0019	0.0019	0.0019	0.0019	0.0019
716	1	0.897	9.31	89.99	0.1917	0.5771	0.10	-0.06	-0.0004	0.18	3.05	0.5562	2.9388	0.0798	0.0030	0.0092	0.010	-0.0004	-0.0000	0.0002	0.2267	0.6338	
	8	0.0780	0.0035	0.0091	0.2284	0.5871	0.0148	0.0006	0.0018	0.0018	0.0018	0.0018	0.0018	0.0018	0.0018	0.0018	0.0018	0.0018	0.0018	0.0018	0.0018	0.0018	0.0018
716	2	0.898	9.30	78.79	0.1892	0.5866	0.10	-0.04	0.19	3.09	0.1457	0.6003	0.6354	0.0796	0.0030	0.0093	0.10	0.04	0.0004	0.0017	0.1457	0.6003	
	8	0.0777	0.0035	0.0087	0.2266	0.5650	0.0318	0.0015	0.0015	0.0015	0.0015	0.0015	0.0015	0.0015	0.0015	0.0015	0.0015	0.0015	0.0015	0.0015	0.0015	0.0015	0.0015
716	3	0.898	9.30	67.49	0.1783	0.6000	0.10	-0.02	0.19	3.12	0.1488	0.6127	0.6101	0.0784	0.0027	0.0083	0.10	0.02	0.0009	0.0023	0.1488	0.6127	
	8	0.0735	0.0035	0.0083	0.2416	0.5695	0.0480	0.0023	0.0023	0.0023	0.0023	0.0023	0.0023	0.0023	0.0023	0.0023	0.0023	0.0023	0.0023	0.0023	0.0023	0.0023	0.0023
716	4	0.898	9.30	56.29	0.1617	0.6060	0.09	0.00	0.18	3.13	0.1483	0.6341	0.5816	0.0746	0.0024	0.0076	0.09	0.00	0.0014	0.0063	0.1483	0.6341	
	8	0.0677	0.0035	0.0076	0.2584	0.5638	0.0603	0.0029	0.0029	0.0029	0.0029	0.0029	0.0029	0.0029	0.0029	0.0029	0.0029	0.0029	0.0029	0.0029	0.0029	0.0029	0.0029
716	5	0.898	9.30	44.99	0.1511	0.6397	0.07	0.02	0.17	3.14	0.1481	0.6148	0.5869	0.0666	0.0020	0.0066	0.07	0.02	0.0019	0.0032	0.1481	0.6148	
	8	0.0576	0.0033	0.0066	0.2872	0.5336	0.0705	0.0032	0.0032	0.0032	0.0032	0.0032	0.0032	0.0032	0.0032	0.0032	0.0032	0.0032	0.0032	0.0032	0.0032	0.0032	0.0032
716	6	0.900	9.30	33.79	0.1457	0.6065	0.05	0.04	0.15	3.15	0.1628	0.5999	0.5555	0.0521	0.0015	0.0055	0.05	0.04	0.0024	0.0087	0.1628	0.5999	
	8	0.0453	0.0028	0.0055	0.3100	0.6065	0.0788	0.0032	0.0032	0.0032	0.0032	0.0032	0.0032	0.0032	0.0032	0.0032	0.0032	0.0032	0.0032	0.0032	0.0032	0.0032	0.0032
716	7	0.897	9.31	22.50	0.1674	0.6564	0.03	0.04	0.12	3.16	0.1777	0.5889	0.5530	0.0336	0.0011	0.0039	0.03	0.04	0.0028	0.0094	0.1777	0.5889	
	8	0.0330	0.0020	0.0039	0.3091	0.6030	0.0856	0.0032	0.0032	0.0032	0.0032	0.0032	0.0032	0.0032	0.0032	0.0032	0.0032	0.0032	0.0032	0.0032	0.0032	0.0032	0.0032
716	8	0.898	9.32	11.30	0.1450	0.6593	0.00	0.05	0.08	3.16	0.1887	0.5826	0.5533	0.0159	0.0011	0.0018	0.00	0.05	0.0030	0.0099	0.1887	0.5826	
	8	0.0170	0.0004	0.0018	0.3680	0.5939	0.0887	0.0031	0.0031	0.0031	0.0031	0.0031	0.0031	0.0031	0.0031	0.0031	0.0031	0.0031	0.0031	0.0031	0.0031	0.0031	0.0031
716	9	0.898	9.32	-0.00	-0.2968	1.4732	0.01	0.06	0.04	3.16	0.1902	0.5774	0.5548	0.0007	-0.0000	0.0002	-0.01	0.06	0.0030	0.0093	0.1902	0.5774	
	8	0.0010	0.0002	-0.0000	-1.0089	-0.2815	0.0926	0.0031	0.0031	0.0031	0.0031	0.0031	0.0031	0.0031	0.0031	0.0031	0.0031	0.0031	0.0031	0.0031	0.0031	0.0031	0.0031
716	10	0.896	9.31	-11.30	0.2573	0.7285	0.05	0.05	0.01	3.16	0.1971	0.5727	0.5620	-0.0145	-0.0007	-0.0008	-0.04	0.05	0.0031	0.0091	0.1971	0.5727	
	8	-0.0140	-0.0008	-0.0020	0.2939	0.7285	0.0944	0.0031	0.0031	0.0031	0.0031	0.0031	0.0031	0.0031	0.0031	0.0031	0.0031	0.0031	0.0031	0.0031	0.0031	0.0031	0.0031

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd																				
716	11	7	0.900	9.31	-22.50	-0.0332	-0.0012	9.31	-0.0012	-0.0039	0.1846	-0.5897	-0.09	0.05	-0.01	3.16	0.2125	0.5746			
		8	-0.0298	-0.0018	-0.0041	-0.0018	-0.0018	0.5897	0.6919	0.0920	0.3127	0.6919	0.09	0.074	0.0031	0.0086	0.1644	0.5746			
		9	-0.0497	-0.0025	-0.0056	-0.0025	-0.0025	-0.11	0.6343	0.0682	0.1897	0.6343	0.04	0.04	-0.0031	3.14	0.2305	0.5749			
716	12	7	0.899	9.30	-33.80	-0.0470	-0.0025	9.30	-0.0025	-0.0056	0.2721	0.6017	-0.11	0.04	0.0027	3.14	0.1529	0.5797			
		8	-0.0639	-0.0022	-0.0076	-0.0022	-0.0022	0.5988	0.5831	0.0606	0.1748	0.5988	0.13	0.03	-0.0029	3.14	0.2440	0.5998			
		9	-0.0571	-0.0031	-0.0066	-0.0031	-0.0031	-0.13	0.5831	0.0831	0.2735	0.5831	0.03	0.0606	0.0022	3.14	0.1354	0.5947			
716	14	7	0.898	9.30	-56.29	-0.0733	-0.0033	9.30	-0.0033	-0.0076	0.1885	-0.5958	-0.14	0.02	-0.007	3.12	0.2590	0.6020			
		8	-0.0664	-0.0033	-0.0076	-0.0033	-0.0033	0.5958	0.5750	0.0493	0.2540	0.5750	0.14	0.02	0.0017	3.12	0.1197	0.6196			
		9	-0.0758	-0.0033	-0.0084	-0.0033	-0.0033	-0.16	0.5552	0.0347	0.2067	0.5896	0.16	0.00	-0.0025	3.11	0.2661	0.6283			
716	15	7	0.898	9.31	-67.49	-0.0809	-0.0032	9.31	-0.0032	-0.0092	0.2077	-0.5825	-0.16	0.00	-0.009	3.11	0.1091	0.6320			
		8	-0.0758	-0.0032	-0.0084	-0.0032	-0.0032	0.5825	0.5552	0.0347	0.2206	0.5896	0.16	0.00	0.0018	3.11	0.2661	0.6320			
		9	-0.0807	-0.0032	-0.0090	-0.0032	-0.0032	-0.16	0.5552	0.0581	0.2031	0.5552	0.16	0.00	0.0012	3.11	0.1091	0.6320			
716	17	7	0.900	9.31	-90.00	-0.0828	-0.0032	9.31	-0.0032	-0.0092	0.2112	-0.5718	-0.17	0.05	-0.009	3.06	0.1801	0.6112			
		8	-0.0807	-0.0032	-0.0092	-0.0032	-0.0032	0.5718	0.5610	0.0217	0.1933	0.5610	0.17	0.05	0.0003	3.06	0.0881	0.6112			
		9	-0.0828	-0.0032	-0.0092	-0.0032	-0.0032	-0.17	0.5610	0.0217	0.2308	0.6032	0.17	0.05	0.0003	3.07	0.3580	0.6504			
717	3	7	0.898	6.22	-89.99	-0.0584	-0.0028	6.22	-0.0028	-0.0074	0.2445	-0.5991	-0.12	0.06	-0.004	3.07	0.0817	0.6504			
		8	-0.0614	-0.0028	-0.0074	-0.0028	-0.0028	0.5991	0.6032	0.0231	0.2308	0.6032	0.12	0.06	0.0003	3.07	0.3580	0.6504			
		9	-0.0591	-0.0028	-0.0072	-0.0028	-0.0028	-0.12	0.6107	0.0351	0.2359	0.6107	0.12	0.04	0.0003	3.09	0.0817	0.6504			
717	4	7	0.898	6.22	-78.79	-0.0574	-0.0028	6.22	-0.0028	-0.0072	0.2359	-0.6023	-0.12	0.04	-0.004	3.09	0.2613	0.6487			
		8	-0.0574	-0.0028	-0.0072	-0.0028	-0.0028	0.6023	0.6107	0.0351	0.2429	0.6107	0.12	0.04	0.0007	3.09	0.1063	0.6487			
		9	-0.0591	-0.0028	-0.0072	-0.0028	-0.0028	-0.12	0.6107	0.0351	0.2429	0.6107	0.12	0.04	0.0007	3.09	0.2613	0.6487			
717	5	7	0.898	6.22	-67.49	-0.0550	-0.0024	6.22	-0.0024	-0.0066	0.2212	-0.6021	-0.11	0.02	-0.003	3.10	0.2606	0.6294			
		8	-0.0550	-0.0024	-0.0066	-0.0024	-0.0024	0.6021	0.5926	0.0473	0.2410	0.5926	0.11	0.02	0.0012	3.10	0.1233	0.6294			
		9	-0.0555	-0.0026	-0.0065	-0.0026	-0.0026	-0.11	0.5926	0.0473	0.2410	0.5926	0.11	0.02	0.0011	3.10	0.2606	0.6294			
717	6	7	0.897	6.21	-56.29	-0.0487	-0.0024	6.21	-0.0024	-0.0058	0.2106	-0.6020	-0.11	0.00	-0.002	3.11	0.2591	0.6280			
		8	-0.0487	-0.0024	-0.0058	-0.0024	-0.0024	0.6020	0.6020	0.0577	0.2541	0.6020	0.11	0.00	0.0016	3.11	0.1359	0.6280			
		9	-0.0487	-0.0024	-0.0058	-0.0024	-0.0024	-0.11	0.6020	0.0577	0.2541	0.6020	0.11	0.00	0.0016	3.11	0.2591	0.6280			

See Key

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key									
717	7	0.897	6.21	-4.99	0.1954	-0.09	0.01	-0.01	3.12	0.2486	0.6218
	8	-0.0433	-0.0016	-0.0056	0.2494	0.6472	0.0451	0.0022	0.0056	0.1498	0.6126
	9	-0.0409	-0.0020	-0.0047		0.5832	0.0653	0.0019	0.0080		
717	8	0.898	6.22	-33.80	0.1827	-0.08	0.02	0.00	3.13	0.2430	0.6090
	8	-0.0335	-0.0012	-0.0042	0.2466	0.6318	0.0511	0.0024	0.0062	0.1684	0.5943
	9	-0.0311	-0.0015	-0.0035		0.5738	0.0697	0.0023	0.0082		
717	9	0.897	6.23	-22.50	0.1876	-0.06	0.02	0.03	3.14	0.2345	0.6030
	8	-0.0220	-0.0008	-0.0027	0.2510	0.6213	0.0560	0.0026	0.0067	0.1647	0.5908
	9	-0.0196	-0.0009	-0.0024		0.6267	0.0726	0.0026	0.0085		
717	10	0.898	6.22	-11.29	0.2177	-0.04	0.02	0.04	3.15	0.2247	0.6010
	8	-0.0100	-0.0004	-0.0012	0.1630	0.6246	0.0585	0.0026	0.0070	0.1914	0.5853
	9	-0.0092	-0.0003	-0.0011		0.5949	0.0739	0.0028	0.0086		
717	11	0.899	6.23	-0.00	0.0171	-0.02	0.02	0.06	3.15	0.2159	0.5999
	8	-0.0001	-0.0000	-0.0000	2.3251	0.6242	0.0595	0.0025	0.0071	0.1975	0.5806
	9	-0.0006	-0.0003	-0.0001		0.8783	0.0739	0.0029	0.0085		
717	12	0.900	6.23	11.29	0.0699	-0.00	0.02	0.09	3.15	0.2034	0.6036
	8	0.0126	0.0001	0.0015	0.4577	0.6319	0.0586	0.0023	0.0070	0.2061	0.5786
	9	0.0106	0.0009	0.0014		0.6758	0.0714	0.0029	0.0082		
717	13	0.899	6.21	22.49	0.1185	0.01	0.01	0.11	3.14	0.1857	0.6154
	8	0.0243	0.0005	0.0031	0.3393	0.6378	0.0564	0.0020	0.0069	0.2131	0.5794
	9	0.0228	0.0015	0.0028		0.6281	0.0678	0.0028	0.0078		
717	14	0.899	6.22	33.79	0.1420	0.03	0.00	0.12	3.14	0.1659	0.6321
	8	0.0359	0.0010	0.0046	0.3220	0.6482	0.0519	0.0017	0.0065	0.2258	0.5903
	9	0.0328	0.0021	0.0042		0.6475	0.0623	0.0020	0.0073		
717	15	0.898	6.19	44.99	0.1663	0.05	-0.00	0.14	3.12	0.1477	0.6399
	8	0.0455	0.0015	0.0059	0.2877	0.6577	0.0442	0.0013	0.0056	0.2310	0.6013
	9	0.0424	0.0024	0.0050		0.5949	0.0559	0.0025	0.0067		
717	16	0.897	6.20	56.29	0.1763	0.06	-0.02	0.16	3.12	0.1336	0.6362
	8	0.0540	0.0019	0.0070	0.2813	0.6561	0.0337	0.0009	0.0042	0.2299	0.6129
	9	0.0493	0.0027	0.0058		0.5939	0.0483	0.0022	0.0059		
717	17	0.898	6.20	67.49	0.1786	0.07	-0.03	0.17	3.10	0.1202	0.6349
	8	0.0594	0.0021	0.0076	0.2854	0.6414	0.0221	0.0005	0.0027	0.2356	0.6236
	9	0.0521	0.0029	0.0064		0.6167	0.0377	0.0017	0.0047		

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																						
717	18	0.898	6.20	78.79	0.1982	0.6258	0.08	-0.05	0.17	3.08	0.2697	0.5921	0.0604	0.0023	0.0075	0.2741	0.6072	0.0081	0.0012	0.0004	0.0032	0.2318	0.6329	
	8	0.0549	0.0030	0.0066	0.0075	0.0066	0.0075	0.0081	0.0012	0.0032	0.0004	0.0032	0.0004	0.0012	0.0032	0.0004	0.0032	0.0004	0.0012	0.0032	0.0004	0.0032	0.0004	0.0032
717	19	0.897	6.20	89.99	0.2196	0.6350	0.08	-0.07	0.17	3.07	-1.1187	-1.3573	0.0593	0.0026	0.0075	0.2414	0.5548	0.0149	-0.0006	-0.0001	0.0018	0.2056	0.6260	
	8	0.0581	0.0028	0.0064	0.0075	0.0064	0.0075	0.0081	0.0012	0.0032	0.0004	0.0032	0.0004	0.0012	0.0032	0.0004	0.0032	0.0004	0.0012	0.0032	0.0004	0.0032	0.0004	0.0032
71A	1	0.899	3.08	89.99	0.2243	0.6583	0.03	-0.06	0.11	3.07	0.6348	1.7003	0.0274	0.0012	0.0036	0.2986	0.6605	0.0167	-0.0005	-0.0001	0.0020	0.1772	0.6234	
	8	0.0286	0.0017	0.0037	0.0036	0.0037	0.0036	0.0042	0.0005	0.0020	0.0001	0.0020	0.0001	0.0005	0.0020	0.0001	0.0020	0.0001	0.0005	0.0020	0.0001	0.0020	0.0001	0.0020
71A	2	0.895	3.09	78.79	0.2226	0.6698	0.03	-0.06	0.11	3.08	0.4506	0.2802	0.0277	0.0012	0.0037	0.3310	0.7150	0.0223	0.0009	0.0002	0.0028	0.2045	0.6315	
	8	0.0277	0.0018	0.0039	0.0037	0.0039	0.0037	0.0042	0.0009	0.0028	0.0002	0.0028	0.0002	0.0009	0.0028	0.0002	0.0028	0.0002	0.0009	0.0028	0.0002	0.0028	0.0002	0.0028
71B	3	0.901	3.09	67.49	0.1947	0.6448	0.02	-0.05	0.11	3.08	0.4187	0.5555	0.0277	0.0010	0.0035	0.1947	0.6257	0.0062	0.0012	0.0006	0.0036	0.2171	0.6374	
	8	0.0285	0.0016	0.0035	0.0035	0.0035	0.0035	0.0042	0.0012	0.0036	0.0006	0.0036	0.0006	0.0012	0.0036	0.0006	0.0036	0.0006	0.0012	0.0036	0.0006	0.0036	0.0006	0.0036
71B	4	0.897	3.09	56.29	0.2058	0.6712	0.02	-0.04	0.11	3.09	0.1912	0.6373	0.0257	0.0010	0.0034	0.3189	0.6423	0.0142	0.0014	0.0004	0.0042	0.2184	0.6373	
	8	0.0250	0.0015	0.0032	0.0032	0.0032	0.0032	0.0036	0.0014	0.0036	0.0004	0.0036	0.0004	0.0014	0.0036	0.0004	0.0036	0.0004	0.0014	0.0036	0.0004	0.0036	0.0004	0.0036
71B	5	0.896	3.09	44.99	0.1467	0.6239	0.02	-0.03	0.10	3.10	0.1836	0.6459	0.0237	0.0006	0.0029	0.3665	0.6590	0.0197	0.0016	0.0004	0.0048	0.2171	0.6369	
	8	0.0200	0.0014	0.0026	0.0026	0.0026	0.0026	0.0032	0.0016	0.0032	0.0004	0.0032	0.0004	0.0016	0.0032	0.0004	0.0032	0.0004	0.0016	0.0032	0.0004	0.0032	0.0004	0.0032
71B	6	0.897	3.10	33.79	0.1695	0.6807	0.01	-0.02	0.09	3.11	0.1947	0.6464	0.0184	0.0006	0.0025	0.4283	0.6890	0.0246	0.0018	0.0003	0.0053	0.2139	0.6364	
	8	0.0146	0.0012	0.0020	0.0020	0.0020	0.0020	0.0026	0.0018	0.0026	0.0003	0.0026	0.0003	0.0018	0.0026	0.0003	0.0026	0.0003	0.0018	0.0026	0.0003	0.0026	0.0003	0.0026
71A	7	0.898	3.10	22.49	0.1083	0.6504	0.00	-0.02	0.08	3.11	0.2015	0.6387	0.0139	0.0003	0.0013	0.4823	0.6915	0.0287	0.0018	0.0003	0.0057	0.2080	0.6332	
	8	0.0093	0.0009	0.0013	0.0013	0.0013	0.0013	0.0016	0.0018	0.0016	0.0003	0.0016	0.0003	0.0018	0.0016	0.0003	0.0016	0.0003	0.0018	0.0016	0.0003	0.0016	0.0003	0.0016
71B	8	0.896	3.10	11.29	0.0872	0.6632	0.00	-0.01	0.07	3.11	0.2122	0.6391	0.0074	0.0001	0.0009	0.7781	0.8613	0.0311	0.0013	0.0003	0.0060	0.2122	0.6391	
	8	0.0039	0.0006	0.0006	0.0006	0.0006	0.0006	0.0007	0.0013	0.0007	0.0003	0.0007	0.0003	0.0013	0.0007	0.0003	0.0007	0.0003	0.0013	0.0007	0.0003	0.0007	0.0003	0.0007
71A	9	0.898	3.10	-0.00	-0.1784	-0.6709	-0.01	-0.01	0.06	3.11	0.2251	0.6445	0.0022	-0.0002	-0.0003	-5.5427	0.4683	0.0325	0.0019	0.0006	0.0062	0.2251	0.6445	
	6	0.0022	-0.0002	-0.0000	-0.0000	-0.0000	-0.0000	-0.0002	0.0014	0.0002	0.0001	0.0002	0.0001	0.0014	0.0002	0.0001	0.0002	0.0001	0.0014	0.0002	0.0001	0.0002	0.0001	0.0002

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key										
718	10	0.899	3.10	-11.30	-0.003	-0.00	0.04	3.11	0.2326	0.6430		
	8	-0.0035	-0.0001	-0.0004	0.2372	0.6384	0.0015	0.0041	0.1891	0.6291		
	9	-0.0048	-0.0000	-0.0006	0.0632	0.6571	0.0018	0.0063				
718	11	0.898	3.09	-22.50	-0.004	-0.00	0.03	3.11	0.2431	0.6400		
	8	-0.0084	-0.0004	-0.0010	0.2766	0.6407	0.0014	0.0039	0.1819	0.6334		
	9	-0.0090	-0.0004	-0.0013	0.2266	0.7578	0.0018	0.0063				
718	12	0.895	3.10	-33.80	-0.005	-0.01	0.02	3.10	0.2512	0.6417		
	8	-0.0139	-0.0007	-0.0019	0.2829	0.6800	0.0013	0.0035	0.1746	0.6410		
	9	-0.0152	-0.0006	-0.0019	0.2106	0.6352	0.0016	0.0062				
718	13	0.895	3.10	-44.99	-0.006	-0.02	0.02	3.10	0.2482	0.6386		
	8	-0.0194	-0.0010	-0.0027	0.2676	0.6967	0.0011	0.0030	0.1528	0.6454		
	9	-0.0204	-0.0008	-0.0024	0.2023	0.5936	0.0014	0.0060				
718	14	0.899	3.09	-56.29	-0.006	-0.03	0.01	3.09	0.2524	0.6339		
	8	-0.0237	-0.0012	-0.0031	0.2696	0.6643	0.0009	0.0023	0.1508	0.6421		
	9	-0.0224	-0.0011	-0.0029	0.2571	0.6606	0.0012	0.0053				
718	15	0.897	3.09	-67.49	-0.007	-0.03	0.01	3.09	0.2549	0.6330		
	8	-0.0288	-0.0013	-0.0035	0.2302	0.6141	0.0006	0.0017	0.1433	0.6563		
	9	-0.0244	-0.0013	-0.0034	0.2828	0.7079	0.0010	0.0048				
718	16	0.898	3.09	-78.79	-0.008	-0.04	0.00	3.07	0.3064	0.5638		
	8	-0.0298	-0.0017	-0.0039	0.2861	0.6653	0.0004	0.0008	0.1332	0.6396		
	9	-0.0289	-0.0012	-0.0034	0.2238	0.5870	0.0008	0.0039				
718	17	0.898	3.10	-89.99	-0.008	-0.06	0.00	3.07	0.3253	0.2681		
	8	-0.0315	-0.0017	-0.0041	0.2815	0.6325	0.0001	0.0001	0.1343	0.6391		
	9	-0.0278	-0.0015	-0.0039	0.2770	0.7090	0.0006	0.0031				
719	1	0.897	-0.000	-90.00	-0.003	-0.06	0.06	3.07	0.2660	0.2018		
	8	-0.0018	-0.0003	-0.0003	0.9895	1.0324	0.0001	0.0001	0.1698	0.6323		
	9	-0.0015	-0.0001	-0.0001	-0.5830	0.5751	0.0008	0.0030				
719	2	0.898	-0.000	-78.80	-0.003	-0.06	0.06	3.07	0.3270	0.3181		
	8	-0.0016	-0.0003	-0.0003	0.9556	0.9800	0.0001	0.0001	0.1693	0.6324		
	9	-0.0000	-0.0000	-0.0003	18.8350-94.1652	0.1652	0.0008	0.0030				
719	3	0.898	-0.000	-67.50	-0.002	-0.05	0.06	3.07	0.3648	0.3742		
	8	-0.0013	-0.0002	-0.0002	0.9877	0.8722	0.0003	0.0003	0.1731	0.6386		
	9	-0.0011	-0.0001	-0.0002	-0.5775	1.0532	0.0003	0.0003				

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																				
		1	2	3	4	5	6	7	8	9	10	11	12	13	14	15	16					
719	4	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9
719	5	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9
719	6	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9
719	7	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9
719	8	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9
719	9	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9
719	10	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9
719	11	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9
719	12	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9
719	13	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9
719	14	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd																									
719 15	7	0.898	-0.001	67.49	-0.01	-0.07	0.06	3.07	0.1595	1.4187																
	8	-0.0020	-0.0001	0.0000	-0.1163	-0.0011	-0.0006	-0.0003	0.1633	0.6495																
	9	-0.0001	0.0002	-0.0000	2.7097	0.0191	0.0006	0.0024																		
719 16	7	0.898	-0.001	78.79	-0.02	-0.06	0.06	3.07	0.2285	1.4094																
	8	-0.0011	-0.0002	0.0000	-0.1208	-0.0014	-0.0000	-0.0003	0.1602	0.6529																
	9	-0.0009	0.0003	-0.0000	0.0655	0.0186	0.0005	0.0024																		
719 17	7	0.898	-0.001	89.99	-0.02	-0.06	0.06	3.07	0.2183	1.1928																
	8	-0.0018	-0.0001	0.0000	-0.1724	-0.0016	-0.0000	-0.0004	0.1620	0.6499																
	9	-0.0009	0.0002	-0.0000	0.1595	0.0182	0.0005	0.0023																		
720 1	7	0.898	-3.11	90.00	-0.07	-0.06	0.01	3.06	-0.0068	-2.4030																
	8	-0.0292	-0.0015	-0.0034	0.5972	0.0004	-0.0000	-0.0001	0.1361	0.6633																
	9	-0.0292	-0.0013	-0.0036	0.6265	0.0193	0.0005	0.0025																		
720 2	7	0.898	-3.11	78.79	-0.08	-0.08	0.01	3.07	0.2519	0.8951																
	8	-0.0281	-0.0015	-0.0034	0.6048	-0.0046	-0.0002	-0.0008	0.1944	0.6461																
	9	-0.0294	-0.0012	-0.0035	0.6009	0.0123	0.0004	0.0015																		
720 3	7	0.898	-3.10	67.50	-0.07	-0.08	0.01	3.06	0.2137	0.7522																
	8	-0.0257	-0.0015	-0.0031	0.6188	-0.0099	-0.0004	-0.0015	0.2204	0.6300																
	9	-0.0269	-0.0011	-0.0034	0.6316	0.0072	0.0003	0.0009																		
720 4	7	0.898	-3.10	56.30	-0.06	-0.08	0.01	3.05	0.2152	0.7080																
	8	-0.0224	-0.0014	-0.0028	0.6323	-0.0144	-0.0006	-0.0020	0.1774	0.6318																
	9	-0.0247	-0.0009	-0.0029	0.5975	0.0034	0.0001	0.0004																		
720 5	7	0.898	-3.10	45.00	-0.06	-0.09	0.02	3.04	0.2193	0.6669																
	8	-0.0191	-0.0011	-0.0023	0.6114	-0.0186	-0.0008	-0.0024	0.3390	1.1243																
	9	-0.0204	-0.0007	-0.0026	0.6495	-0.0008	-0.0000	-0.0001	0.2251	0.6680																
720 6	7	0.899	-3.10	33.80	-0.05	-0.10	0.03	3.04	0.2251	0.6214																
	8	-0.0142	-0.0009	-0.0017	0.6209	-0.0217	-0.0009	-0.0029	0.2228	0.6650																
	9	-0.0154	-0.0004	-0.0020	0.6620	-0.0035	-0.0002	-0.0005	0.2678	0.7138																
720 7	7	0.897	-3.10	22.50	-0.04	-0.10	0.04	3.04	0.2187	0.6501																
	8	-0.0083	-0.0006	-0.0010	0.6376	-0.0244	-0.0010	-0.0032	0.3105	0.7317																
	9	-0.0100	-0.0002	-0.0011	0.5897	-0.0056	-0.0003	-0.0008																		
720 8	7	0.898	-3.09	11.29	-0.03	-0.10	0.05	3.03	0.2187	0.6501																
	8	-0.0034	-0.0003	-0.0004	0.6231	-0.0261	-0.0011	-0.0034	0.3105	0.7317																
	9	-0.0049	-0.0000	-0.0006	0.6558	-0.0066	-0.0004	-0.0009																		

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key												
720 9	7	0.900	-3.10	-0.000	-0.000	-0.03	-0.10	0.06	3.03	0.2142	0.6593			
	6	0.0012	-0.0001	0.0002	1.0354	-0.0265	-0.0011	-0.0035	0.2979	0.6640				
	9	-0.0009	0.0002	-0.0001	0.9943	-0.0070	-0.0004	-0.0009						
720 10	7	0.897	-3.10	-11.30	-0.02	-0.10	0.07	3.03	0.1971	0.6522				
	6	0.0057	0.0001	0.0008	0.7145	-0.0264	-0.0010	-0.0034	0.3254	0.6995				
	9	0.0028	0.0004	0.0003	0.5695	-0.0062	-0.0004	-0.0008						
720 11	7	0.897	-3.10	-22.50	-0.00	-0.10	0.07	3.03	0.1829	0.6553				
	6	0.0116	0.0002	0.0013	0.5911	-0.0248	-0.0009	-0.0032	0.3679	0.7529				
	9	0.0066	0.0007	0.0009	0.7171	-0.0047	-0.0003	-0.0007						
720 12	7	0.897	-3.10	-33.80	0.00	-0.10	0.08	3.04	0.1650	0.6666				
	6	0.0155	0.0006	0.0020	0.6559	-0.0217	-0.0007	-0.0029	0.4797	0.8120				
	9	0.0134	0.0008	0.0014	0.5242	-0.0023	-0.0002	-0.0003						
720 13	7	0.898	-3.10	-45.00	0.01	-0.09	0.09	3.04	0.1471	0.6834				
	6	0.0198	0.0008	0.0024	0.6204	-0.0179	-0.0005	-0.0024	-0.4214	0.5081				
	9	0.0178	0.0010	0.0019	0.5505	-0.0007	-0.0000	-0.0000						
720 14	7	0.897	-3.10	-56.30	0.01	-0.07	0.10	3.05	0.1227	0.7100				
	6	0.0235	0.0009	0.0028	0.6141	-0.0131	-0.0003	-0.0018	0.1535	0.6627				
	9	0.0217	0.0012	0.0025	0.5822	-0.0044	-0.0001	-0.0005						
720 15	7	0.899	-3.10	-67.50	0.02	-0.07	0.10	3.06	0.1797	0.7559				
	6	0.0262	0.0010	0.0032	0.6114	-0.0065	-0.0002	-0.0009	0.2011	0.6753				
	9	0.0245	0.0014	0.0029	0.6010	-0.0091	-0.0003	-0.0012						
720 16	7	0.897	-3.11	-78.80	0.02	-0.06	0.11	3.07	0.2847	1.2795				
	6	0.0279	0.0010	0.0033	0.6052	-0.0113	-0.0005	-0.0020	0.1919	0.6569				
	9	0.0250	0.0015	0.0033	0.6630	-0.0155	-0.0005	-0.0020						
720 17	7	0.897	-3.11	-90.00	0.02	-0.06	0.11	3.07	0.4967	0.2415				
	6	0.0285	0.0010	0.0034	0.5956	0.0020	0.0002	0.0000	0.1938	0.6287				
	9	0.0270	0.0014	0.0032	0.5928	0.0215	0.0006	0.0027						
721 1	7	0.896	-3.10	-90.00	0.01	-0.06	0.12	6.05	0.5156	0.2821				
	6	0.0271	0.0010	0.0033	0.6255	0.0021	0.0002	0.0001	0.2174	0.6321				
	9	0.0273	0.0015	0.0033	0.6207	0.0398	0.0017	0.0050						
721 2	7	0.897	-3.10	-78.80	0.01	-0.06	0.11	6.05	0.3211	1.5214				
	6	0.0268	0.0011	0.0033	0.6209	-0.0010	-0.0000	-0.0003	0.2262	0.6440				
	9	0.0281	0.0014	0.0032	0.5738	-0.0333	-0.0015	-0.0042						

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																			
721	3	0.896	-3.10	-67.50	0.2112	0.01	-0.09	0.11	-0.0009	0.2002	0.7790	0.0253	0.0010	0.0031	0.2686	0.5284	-0.057	-0.0012	0.0009	0.2295	0.6561
	8	0.0260	0.0013	0.0030	0.0031	0.5880	0.0274	0.0012	0.0035												
	9																				
721	4	0.899	-3.10	-56.30	0.1987	0.01	-0.08	0.10	-0.003	0.1206	0.7173	0.0230	0.0009	0.0027	0.3072	0.6011	-0.025	-0.0009	0.0017	0.2166	0.6494
	8	0.0218	0.0013	0.0027	0.0027	0.6256	0.0221	0.0009	0.0028												
	9																				
721	5	0.899	-3.09	-45.00	0.1834	0.01	-0.09	0.09	-0.0024	0.1474	0.6952	0.0196	0.0007	0.0023	0.3017	0.5992	-0.0174	-0.0007	0.0023	0.2166	0.6783
	8	0.0186	0.0011	0.0021	0.0021	0.5686	0.0170	0.0007	0.0023												
	9																				
721	6	0.898	-3.09	-33.80	0.2338	0.00	-0.10	0.09	-0.0028	0.1626	0.6630	0.0146	0.0006	0.0019	0.2338	0.6672	-0.0215	-0.0005	0.0028	0.2048	0.6781
	8	0.0144	0.0008	0.0015	0.0015	0.5205	0.0135	0.0005	0.0018												
	9																				
721	7	0.898	-3.09	-22.50	0.1404	0.00	-0.10	0.08	-0.0031	0.1846	0.6619	0.0108	0.0003	0.0013	0.1404	0.6162	-0.0241	-0.0004	0.0031	0.2374	0.7123
	8	0.0090	0.0006	0.0009	0.0009	0.5166	0.0101	0.0004	0.0014												
	9																				
721	8	0.897	-3.12	-11.30	0.0877	0.00	-0.10	0.07	-0.0012	0.1995	0.6567	0.0052	0.0005	0.0004	0.0877	0.7805	-0.0257	-0.0004	0.0033	0.2488	0.7164
	8	0.0029	0.0005	0.0004	0.0004	0.7805	0.0083	0.0004	0.0012												
	9																				
721	9	0.899	-3.10	-0.00	0.4085	0.00	-0.10	0.06	-0.0011	0.2180	0.6592	0.0006	0.0002	0.0001	-1.4085	1.3200	-0.0260	-0.0003	0.0011	0.2527	0.7230
	8	0.0004	0.0002	0.0001	0.0001	1.3200	0.0077	0.0003	0.0011												
	9																				
721	10	0.897	-3.11	11.29	0.4453	0.03	-0.10	0.05	-0.0034	0.2246	0.6651	0.0043	0.0003	0.0005	0.4453	0.6199	-0.0255	-0.0004	0.0034	0.2507	0.7251
	8	0.0041	0.0000	0.0006	0.0006	0.7419	0.0080	0.0004	0.0011												
	9																				
721	11	0.898	-3.09	22.49	0.3471	0.04	-0.10	0.04	-0.0013	0.2353	0.6625	0.0094	0.0006	0.0010	0.3471	0.6175	-0.0240	-0.0004	0.0013	0.2339	0.6765
	8	0.0092	0.0002	0.0010	0.0010	0.5866	0.0096	0.0004	0.0013												
	9																				
721	12	0.898	-3.09	33.80	0.3029	0.04	-0.10	0.03	-0.0028	0.2251	0.6663	0.0151	0.0009	0.0018	0.3029	0.6010	-0.0212	-0.0004	0.0028	0.1685	0.6877
	8	0.0132	0.0005	0.0019	0.0019	0.7477	0.0134	0.0004	0.0018												
	9																				
721	13	0.896	-3.10	45.00	0.2670	0.05	-0.09	0.02	-0.0023	0.2315	0.6906	0.0203	0.0010	0.0025	0.2670	0.5816	-0.0177	-0.0005	0.0023	0.1614	0.6860
	8	0.0187	0.0007	0.0025	0.0025	0.6697	0.0174	0.0005	0.0023												
	9																				

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																		
721	14	7	0.896	-3.11	56.30	0.3010	-0.06	-0.09	0.02	6.03	0.2124	0.6942	-0.0234	-0.0014	-0.0029	0.6247	-0.0140	-0.0005	-0.0019	0.1619	0.6719
		8	-0.0228	-0.0009	-0.0028	0.2082	0.6319	0.0224	0.0007	0.0030											
		9																			
721	15	7	0.898	-3.12	67.50	0.2818	-0.07	-0.08	0.01	6.03	0.2068	0.7480	-0.0267	-0.0015	-0.0032	0.6155	-0.0094	-0.0003	-0.0014	0.1544	0.6652
		8	-0.0267	-0.0015	-0.0032	0.2067	0.5990	0.0281	0.0008	0.0037											
		9																			
721	16	7	0.899	-3.12	78.80	0.2444	-0.07	-0.07	0.01	6.04	0.2944	0.9905	-0.0273	-0.0012	-0.0034	0.5918	-0.0038	-0.0002	-0.0007	0.1459	0.6580
		8	-0.0297	-0.0014	-0.0035	0.2339	0.6277	0.0345	0.0010	0.0045											
		9																			
721	17	7	0.899	-3.12	90.00	0.2404	-0.07	-0.06	0.01	6.05	0.2011	-0.0045	-0.0304	-0.0013	-0.0036	0.5991	0.0007	0.0000	-0.0001	0.2011	0.6504
		8	-0.0304	-0.0014	-0.0036	0.2389	0.6261	0.0407	0.0011	0.0053											
		9																			
722	1	7	0.896	-0.003	89.99	0.7176	-0.03	-0.06	0.06	6.05	0.2602	1.5514	-0.0019	-0.0002	-0.0001	0.4042	-0.0011	-0.0000	-0.0003	0.2602	0.6499
		8	-0.0019	-0.0002	-0.0001	0.2387	0.5434	0.0368	0.0015	0.0047											
		9																			
722	2	7	0.897	-0.003	78.79	0.3190	-0.03	-0.06	0.06	6.06	0.2178	1.4108	-0.0028	-0.0001	-0.0001	0.2533	-0.0011	-0.0000	-0.0003	0.2178	0.6589
		8	-0.0028	-0.0001	-0.0001	0.3237	0.6919	0.0370	0.0015	0.0048											
		9																			
722	3	7	0.896	-0.003	67.49	0.6141	-0.03	-0.06	0.06	6.05	0.0196	1.5653	-0.0020	-0.0002	-0.0000	0.4187	-0.0007	-0.0000	-0.0002	0.0196	0.6553
		8	-0.0020	-0.0002	-0.0000	0.6158	0.7658	0.0374	0.0015	0.0049											
		9																			
722	4	7	0.896	-0.002	56.29	1.0286	-0.02	-0.06	0.06	6.05	-1.2802	6.5578	-0.0013	-0.0002	-0.0000	0.4227	-0.0001	-0.0000	-0.0001	-1.2802	0.6558
		8	-0.0013	-0.0002	-0.0000	0.9338	0.9992	0.0377	0.0015	0.0049											
		9																			
722	5	7	0.897	-0.002	44.99	0.8025	-0.02	-0.06	0.06	6.05	0.5966	-0.1863	-0.0010	-0.0001	-0.0000	0.3420	0.0006	0.0000	-0.0000	0.5966	0.6559
		8	-0.0010	-0.0001	-0.0000	0.7689	0.0952	0.0377	0.0015	0.0049											
		9																			
722	6	7	0.897	-0.002	33.79	-6.0959	-0.02	-0.06	0.06	6.05	0.5481	0.3817	0.0001	-0.0001	-0.0000	0.4181	0.0001	0.0000	0.0000	0.5481	0.6577
		8	-0.0001	-0.0001	-0.0000	0.4814	0.0999	0.0377	0.0015	0.0049											
		9																			
722	7	7	0.896	-0.001	22.49	-0.7296	-0.02	-0.06	0.06	6.05	0.4526	0.4335	0.0009	-0.0001	-0.0000	1.4405	0.0020	0.0001	0.0000	0.4526	0.6602
		8	-0.0010	-0.0001	-0.0000	0.3861	0.0067	0.0376	0.0015	0.0049											
		9																			

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																					
722	8	0.897	-0.001	11.29	-0.3873	-0.02	-0.06	0.06	0.002	6.05	0.4323	0.4726	0.0016	-0.0001	0.0003	0.0003	1.0902	0.0024	0.0002	0.0002	0.0002	0.2102	0.6571
	8	0.0002	0.0002	0.0000	-4.5555	0.2683	0.0376	0.0015	0.0015	0.0049													
722	9	0.898	-0.001	-0.00	-0.3925	-0.02	-0.05	0.06	0.002	6.05	0.4272	0.4782	0.0020	-0.0001	0.0003	0.0003	0.8475	0.0026	0.0002	0.0002	0.0002	0.2094	0.6575
	8	0.0004	0.0002	0.0000	-3.4029	-0.2544	0.0375	0.0015	0.0015	0.0049													
722	10	0.898	-0.001	-11.30	-0.4866	-0.02	-0.05	0.06	0.002	6.05	0.4239	0.4692	0.0018	-0.0001	0.0002	0.0002	0.7105	0.0027	0.0002	0.0002	0.0002	0.2077	0.6556
	8	0.0002	0.0002	-0.0000	-5.1153	1.9992	0.0375	0.0015	0.0015	0.0049													
722	11	0.896	-0.001	-22.50	-0.7157	-0.02	-0.05	0.06	0.002	6.05	0.3995	0.4685	0.0005	-0.0000	0.0002	0.0002	2.4767	0.0029	0.0002	0.0002	0.0002	0.2050	0.6481
	8	0.0005	0.0001	0.0002	-0.7157	74.8204	0.0381	0.0015	0.0015	0.0049													
722	12	0.897	-0.001	-33.80	-2.8514	0.1252	-0.06	0.05	0.002	6.05	0.3988	0.4698	0.0004	-0.0002	0.0000	0.0000	0.1252	0.0029	0.0002	0.0002	0.0002	0.2056	0.6442
	8	0.0006	0.0001	-0.0001	-1.5075	0.8003	0.0389	0.0016	0.0016	0.0050													
722	13	0.897	-0.001	-45.00	0.7023	-0.03	-0.05	0.05	0.002	6.05	0.3775	0.4580	0.0012	-0.0001	0.0001	0.0001	0.4445	0.0031	0.0002	0.0002	0.0002	0.2007	0.6423
	8	0.0002	0.0001	-0.0002	2.9937	-5.3689	0.0399	0.0016	0.0016	0.0051													
722	14	0.896	-0.001	-56.30	0.6141	-0.03	-0.05	0.05	0.002	6.05	0.3764	0.4619	0.0018	-0.0002	0.0002	0.0002	0.6994	0.0032	0.0002	0.0002	0.0002	0.2007	0.6432
	8	0.0015	0.0000	-0.0003	0.1003	-1.1414	0.0412	0.0016	0.0016	0.0053													
722	15	0.897	-0.001	-67.50	0.5690	-0.03	-0.05	0.06	0.002	6.05	0.3420	0.4216	0.0023	-0.0002	0.0003	0.0003	0.7463	0.0033	0.0002	0.0002	0.0002	0.1976	0.6399
	8	0.0003	0.0001	-0.0001	-2.2716	2.9596	0.0422	0.0016	0.0016	0.0054													
722	16	0.897	-0.001	-78.80	0.5684	-0.03	-0.05	0.06	0.002	6.06	0.3605	0.4236	0.0026	-0.0001	0.0004	0.0004	0.2033	0.0029	0.0002	0.0002	0.0002	0.1961	0.6382
	8	0.0000	0.0001	-0.0001	12.3436	-13.5870	0.0426	0.0016	0.0016	0.0054													
722	17	0.897	-0.001	-90.00	0.5646	-0.03	-0.06	0.06	0.002	6.06	0.3883	0.3621	0.0028	-0.0003	0.0004	0.0004	0.8009	0.0027	0.0002	0.0002	0.0002	0.1938	0.6389
	8	0.0001	0.0001	-0.0001	-9.4244	5.3045	0.0431	0.0016	0.0016	0.0054													
723	1	0.897	-0.001	-89.99	0.2522	-0.08	-0.05	0.01	0.002	6.05	0.3838	0.4160	0.0031	-0.0014	0.0042	0.0042	0.6426	0.0027	0.0002	0.0002	0.0002	0.1485	0.6436
	8	0.0016	-0.0016	-0.0042	0.2522	0.6443	0.0450	0.0013	0.0013	0.0058													

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key →														
723	2	7	0.097	3.09	-76.79	-0.09	-0.05	0.00	6.06	0.2993	0.5813	-0.0309	-0.0016	0.0004	0.0009	0.1595	0.6393
		8	-0.0273	-0.0012	-0.0032	0.2679	0.6781	0.0078	0.0004	0.0064							
		9	-0.0245	-0.0012	-0.0030	0.2317	0.5981	0.0507	0.0016	0.0064							
723	3	7	0.097	3.08	-67.49	-0.07	-0.04	0.01	6.07	0.2530	0.6437	-0.0289	-0.0018	0.0006	0.0017	0.1632	0.6310
		8	-0.0289	-0.0014	-0.0038	0.2426	0.6580	0.0138	0.0006	0.0017							
		9	-0.0245	-0.0012	-0.0030	0.2479	0.6307	0.0554	0.0018	0.0017							
723	4	7	0.097	3.08	-56.29	-0.06	-0.03	0.01	6.07	0.2489	0.6396	-0.0259	-0.0011	0.0009	0.0024	0.1738	0.6138
		8	-0.0259	-0.0010	-0.0032	0.1944	0.6170	0.0193	0.0009	0.0024							
		9	-0.0205	-0.0011	-0.0029	0.2759	0.7075	0.0583	0.0020	0.0071							
723	5	7	0.097	3.08	-44.99	-0.06	-0.02	0.03	6.08	0.2459	0.6339	-0.0199	-0.0026	0.0012	0.0031	0.1775	0.6017
		8	-0.0199	-0.0009	-0.0026	0.2300	0.6649	0.0244	0.0012	0.0031							
		9	-0.0170	-0.0008	-0.0024	0.2601	0.7179	0.0612	0.0021	0.0073							
723	6	7	0.095	3.09	-33.79	-0.05	-0.01	0.03	6.08	0.2432	0.6382	-0.0143	-0.0019	0.0013	0.0036	0.1875	0.6008
		8	-0.0143	-0.0007	-0.0019	0.2541	0.6920	0.0283	0.0013	0.0036							
		9	-0.0133	-0.0006	-0.0019	0.2303	0.7145	0.0628	0.0023	0.0075							
723	7	7	0.097	3.10	-22.50	-0.04	-0.01	0.04	6.08	0.2337	0.6368	-0.0088	-0.0011	0.0014	0.0040	0.1939	0.5951
		8	-0.0088	-0.0004	-0.0011	0.2404	0.6651	0.0314	0.0014	0.0040							
		9	-0.0076	-0.0003	-0.0012	0.2267	0.8132	0.0637	0.0024	0.0075							
723	8	7	0.096	3.09	-11.30	-0.03	-0.01	0.06	6.09	0.2304	0.6433	-0.0037	-0.0005	0.0015	0.0042	0.2002	0.5937
		8	-0.0037	-0.0001	-0.0005	0.2555	0.7456	0.0329	0.0015	0.0042							
		9	-0.0036	-0.0000	-0.0005	0.0557	0.8053	0.0639	0.0025	0.0075							
723	9	7	0.095	3.09	-0.00	-0.02	-0.01	0.07	6.09	0.2178	0.6454	-0.0011	0.0003	0.0014	0.0042	0.2047	0.5959
		8	-0.0011	0.0000	0.0002	0.2710	1.1731	0.0332	0.0014	0.0042							
		9	-0.0000	0.0003	0.0001-23.1317	-9.6582	0.0633	0.0025	0.0025	0.0075							
723	10	7	0.096	3.10	11.29	-0.00	-0.01	0.08	6.09	0.2045	0.6399	-0.0073	-0.0006	0.0013	0.0041	0.2104	0.5989
		8	-0.0073	0.0001	0.0009	0.0902	0.6388	0.0320	0.0013	0.0041							
		9	-0.0064	0.0006	0.0006	0.4823	0.5223	0.0618	0.0026	0.0074							
723	11	7	0.097	3.09	22.49	0.00	-0.02	0.09	6.08	0.1950	0.6445	-0.0134	0.0002	0.0011	0.0037	0.2088	0.5979
		8	-0.0134	0.0002	0.0017	0.0966	0.6436	0.0292	0.0011	0.0037							
		9	-0.0111	0.0009	0.0013	0.4248	0.6158	0.0603	0.0025	0.0072							
723	12	7	0.098	3.08	33.79	0.01	-0.03	0.09	6.08	0.1839	0.6409	-0.0186	0.0005	0.0009	0.0032	0.2122	0.6019
		8	-0.0186	0.0005	0.0023	0.1387	0.6391	0.0251	0.0009	0.0032							
		9	-0.0159	0.0013	0.0021	0.4075	0.6615	0.0576	0.0024	0.0069							

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key											
723	13	7	0.897	3.08	44.99	0.1847	0.02	-0.04	0.11	6.08	0.1750	0.6449		
		8	0.0222	0.0008	0.0029	0.3566	0.6703	0.0201	0.0007	0.0026	0.2174	0.6121		
		9	0.0212	0.0015	0.0027	0.0000	0.6553	0.0543	0.0023	0.0066	0.0000	0.0000		
723	14	7	0.898	3.08	56.29	0.1850	0.02	-0.04	0.11	6.07	0.1766	0.6460		
		8	0.0255	0.0009	0.0033	0.3341	0.6578	0.0147	0.0005	0.0019	0.2153	0.6134		
		9	0.0254	0.0016	0.0033	0.0000	0.6672	0.0511	0.0022	0.0062	0.0000	0.0000		
723	15	7	0.897	3.07	67.49	0.3052	0.03	-0.05	0.12	6.06	0.3853	0.5723		
		8	0.0270	0.0011	0.0035	0.3221	0.6593	0.0065	0.0005	0.0007	0.2188	0.6228		
		9	0.0283	0.0018	0.0038	0.0000	0.6719	0.0464	0.0020	0.0057	0.0000	0.0000		
723	16	7	0.897	3.08	78.79	0.2033	0.03	-0.05	0.12	6.06	0.4099	0.3896		
		8	0.0276	0.0011	0.0035	0.3022	0.6472	0.0026	0.0002	0.0002	0.2231	0.6334		
		9	0.0301	0.0018	0.0039	0.0000	0.6548	0.0407	0.0018	0.0051	0.0000	0.0000		
723	17	7	0.897	3.07	89.99	0.2035	0.02	-0.07	0.12	6.05	0.8946	1.9884		
		8	0.0272	0.0011	0.0034	0.3110	0.6400	0.0006	-0.0001	-0.0002	0.2247	0.6375		
		9	0.0290	0.0018	0.0039	0.0000	0.6851	0.0346	0.0015	0.0044	0.0000	0.0000		
724	1	7	0.899	6.19	89.99	0.2128	0.08	-0.06	0.17	6.05	-0.4155	-0.3927		
		8	0.0590	0.0025	0.0073	0.2677	0.6228	0.0008	-0.0000	-0.0000	0.2339	0.6214		
		9	0.0560	0.0030	0.0058	0.0000	0.6068	0.0323	0.0015	0.0041	0.2427	0.6203		
724	2	7	0.899	6.19	78.79	0.2063	0.08	-0.04	0.17	6.06	0.2339	0.6214		
		8	0.0590	0.0024	0.0075	0.2635	0.6383	0.0090	0.0004	0.0011	0.2339	0.6214		
		9	0.0565	0.0029	0.0066	0.0000	0.5849	0.0436	0.0021	0.0034	0.2339	0.6214		
724	3	7	0.898	6.19	67.49	0.1831	0.07	-0.03	0.17	6.08	0.1187	0.6374		
		8	0.0581	0.0021	0.0075	0.2761	0.6454	0.0228	0.0005	0.0029	0.2393	0.5992		
		9	0.0536	0.0029	0.0063	0.0000	0.5954	0.0531	0.0025	0.0063	0.2393	0.5992		
724	4	7	0.900	6.19	56.29	0.1816	0.06	-0.01	0.16	6.09	0.1361	0.6440		
		8	0.0523	0.0019	0.0068	0.2705	0.6581	0.0340	0.0009	0.0043	0.2355	0.5828		
		9	0.0504	0.0027	0.0057	0.0000	0.5715	0.0613	0.0028	0.0071	0.2355	0.5828		
724	5	7	0.900	6.19	44.99	0.1610	0.05	-0.00	0.15	6.10	0.1495	0.6466		
		8	0.0449	0.0014	0.0058	0.2840	0.6480	0.0445	0.0013	0.0057	0.2279	0.5707		
		9	0.0432	0.0024	0.0050	0.0000	0.5861	0.0681	0.0031	0.0077	0.2279	0.5707		
724	6	7	0.899	6.19	33.79	0.1470	0.03	0.01	0.14	6.10	0.1673	0.6365		
		8	0.0351	0.0010	0.0045	0.3034	0.6443	0.0520	0.0017	0.0066	0.2169	0.5641		
		9	0.0342	0.0020	0.0041	0.0000	0.6074	0.0735	0.0031	0.0083	0.2169	0.5641		

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd																				
724	7	0.899	6.20	22.49	0.1096	0.01	0.02	0.12	6.11	0.1846	0.6199	0.0248	0.0005	0.0029	0.3349	0.6000	0.0565	0.0020	0.0070	0.2026	0.5592
	8	0.0239	0.0016	0.0029	0.0910	0.6184	0.0780	0.0031	0.0087												
	9	0.901	6.21	11.29	0.4382	0.6382	0.0815	0.10	6.11	0.2013	0.6089	0.0125	0.0002	0.0015	0.0910	0.6269	0.0586	0.0023	0.0071	0.1934	0.5583
724	7	0.898	6.21	-0.00	0.2374	-0.01	0.03	0.07	6.11	0.2139	0.6044	0.0003	0.0000	0.0000	1.3070	-0.0630	0.0596	0.0025	0.0072	0.1869	0.5604
	8	0.0014	0.0003	0.0001	0.2583	0.6619	0.0839	0.0031	0.0094												
	9	0.898	6.21	-11.29	0.2094	0.7345	0.0843	0.04	6.11	0.2237	0.6045	-0.0095	-0.0004	-0.0012	0.2583	0.6670	0.0588	0.0026	0.0071	0.1802	0.5578
724	7	0.898	6.21	-22.50	0.1997	0.6408	0.0562	0.02	6.11	0.2315	0.6034	-0.0074	-0.0009	0.2684	0.6800	0.0833	0.0029	0.0094	0.1791	0.5657	
	8	0.0216	-0.0008	-0.0027	0.1775	0.6345	0.0513	0.01	6.11	0.2408	0.6112	-0.0179	-0.0015	0.2855	0.6605	0.0815	0.0027	0.0093	0.1712	0.5717	
	9	0.0336	-0.0011	-0.0042	0.1873	0.6471	0.0452	0.0022	6.09	0.2447	0.6223	-0.0377	-0.0020	0.2737	0.6359	0.0779	0.0025	0.0091	0.1619	0.5850	
724	7	0.898	6.20	-44.99	0.1952	0.6388	0.0357	-0.02	6.08	0.2575	0.6315	-0.0519	-0.0020	0.1952	0.6301	0.0722	0.0021	0.0086	0.2575	0.6315	
	8	0.0434	-0.0016	-0.0056	0.2675	0.6301	0.0722	0.0021	6.07	0.2567	0.6303	-0.0461	-0.0024	0.2675	0.6301	0.0722	0.0021	0.0086	0.1472	0.5953	
	9	0.898	6.20	-67.49	0.2113	0.6331	0.0252	0.0017	6.07	0.2567	0.6303	-0.0521	-0.0027	0.2651	0.6331	0.0646	0.0017	0.0078	0.1342	0.6099	
724	7	0.897	6.20	-78.79	0.2170	0.5979	0.0150	-0.03	6.06	0.2531	0.6177	-0.0591	-0.0025	0.2170	0.5979	0.0150	0.0007	0.0018	0.2531	0.6177	
	8	0.0571	-0.0028	-0.0071	0.2466	0.6116	0.0440	-0.0003	6.04	0.2557	0.6420	-0.0571	-0.0028	0.2501	0.6210	0.0551	0.0014	0.0070	0.1334	0.6363	
	9	0.899	6.20	-89.99	0.2338	0.6029	0.0028	-0.0072	6.04	0.2557	0.6420	-0.0601	-0.0028	0.2338	0.6029	0.0028	0.0009	0.0056	0.1054	0.6420	

See Key →

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key															
725	1	7	0.898	9.32	-90.00	-0.0033	-0.0094	-0.0091	0.2065	0.1990	-0.16	-0.06	-0.07	6.04	0.1965	0.3682	
		6	-0.0820	-0.0032	-0.0094	0.0033	-0.0094	-0.0091	0.2065	0.1990	0.5761	0.0029	0.0001	0.0002	0.0929	0.6280	
		9	-0.0812	-0.0032	-0.0091	-0.0032	-0.0091	-0.0091	0.1990	0.5637	0.0429	0.0007	0.0007	0.0053			
725	2	7	0.897	9.33	-78.79	-0.0031	-0.0096	-0.0089	0.1904	0.2143	-0.15	-0.03	-0.07	6.06	0.2593	0.6314	
		6	-0.0834	-0.0031	-0.0096	0.0031	-0.0096	-0.0089	0.1904	0.2143	0.5765	0.0186	0.0009	0.0023	0.1016	0.6342	
		9	-0.0784	-0.0033	-0.0089	-0.0033	-0.0089	-0.0089	0.2143	0.5696	0.0612	0.0012	0.0012	0.0077			
725	3	7	0.899	9.33	-67.49	-0.0031	-0.0096	-0.0082	0.1938	0.2276	-0.15	-0.00	-0.06	6.08	0.2705	0.6343	
		6	-0.0803	-0.0031	-0.0096	0.0031	-0.0096	-0.0082	0.1938	0.2276	0.5999	0.0344	0.0018	0.0043	0.1109	0.6111	
		9	-0.0738	-0.0033	-0.0082	-0.0033	-0.0082	-0.0082	0.2276	0.5576	0.0755	0.0016	0.0016	0.0092			
725	4	7	0.900	9.32	-56.29	-0.0027	-0.0073	-0.0073	0.1848	0.2497	-0.14	0.02	-0.05	6.09	0.2564	0.6036	
		6	-0.0737	-0.0027	-0.0073	0.0027	-0.0073	-0.0073	0.1848	0.2497	0.6092	0.0495	0.0025	0.0059	0.1286	0.5912	
		9	-0.0659	-0.0032	-0.0073	-0.0032	-0.0073	-0.0073	0.2497	0.5554	0.0865	0.0022	0.0022	0.0102			
725	5	7	0.898	9.32	-44.99	-0.0023	-0.0078	-0.0065	0.1827	0.2849	-0.12	0.04	-0.04	6.11	0.2451	0.5926	
		6	-0.0631	-0.0023	-0.0078	0.0023	-0.0078	-0.0065	0.1827	0.2849	0.6239	0.0607	0.0029	0.0072	0.1479	0.5729	
		9	-0.0551	-0.0031	-0.0065	-0.0031	-0.0065	-0.0065	0.2849	0.5935	0.0937	0.0027	0.0027	0.0107			
725	6	7	0.898	9.31	-33.79	-0.0018	-0.0055	-0.0055	0.1845	0.2842	-0.10	0.05	-0.02	6.12	0.2307	0.5779	
		6	-0.0493	-0.0018	-0.0055	0.0018	-0.0055	-0.0055	0.1845	0.2842	0.6366	0.0685	0.0031	0.0079	0.1512	0.5643	
		9	-0.0446	-0.0025	-0.0055	-0.0025	-0.0055	-0.0055	0.2842	0.6236	0.0988	0.0029	0.0029	0.0111			
725	7	7	0.898	9.32	-22.50	-0.0012	-0.0036	-0.0036	0.2056	0.2726	-0.07	0.06	0.00	6.12	0.2113	0.5734	
		6	-0.0311	-0.0012	-0.0036	0.0012	-0.0036	-0.0036	0.2056	0.2726	0.6112	0.0753	0.0031	0.0086	0.1484	0.5520	
		9	-0.0297	-0.0016	-0.0036	-0.0016	-0.0036	-0.0036	0.2726	0.6149	0.1007	0.0029	0.0029	0.0111			
725	8	7	0.897	9.32	-11.30	-0.0007	-0.0018	-0.0018	0.2709	0.2906	-0.05	0.06	0.03	6.12	0.1980	0.5753	
		6	-0.0134	-0.0007	-0.0018	0.0007	-0.0018	-0.0018	0.2709	0.2906	0.5548	0.0795	0.0031	0.0091	0.1433	0.5490	
		9	-0.0127	-0.0007	-0.0018	-0.0007	-0.0018	-0.0018	0.2906	0.7218	0.1013	0.0029	0.0029	0.0111			
725	9	7	0.898	9.32	-0.00	-0.0003	-0.0001	-0.0001	-0.1738	-2.2981	-0.02	0.05	0.07	6.11	0.1887	0.5759	
		6	-0.0024	-0.0003	-0.0003	0.0003	-0.0001	-0.0001	-0.1738	-2.2981	0.7213	0.0619	0.0030	0.0094	0.1451	0.5545	
		9	0.0007	0.0003	0.0001	0.0003	0.0001	0.0001	-2.2981	1.2823	0.0992	0.0028	0.0028	0.0110			
725	10	7	0.899	9.33	11.30	0.0005	0.0020	0.0020	0.1469	0.3927	0.01	0.05	0.10	6.11	0.1854	0.5817	
		6	-0.0179	0.0005	0.0020	0.0005	0.0020	0.0020	0.1469	0.3927	0.6068	0.0616	0.0030	0.0035	0.1491	0.5520	
		9	0.0161	0.0012	0.0020	0.0012	0.0020	0.0020	0.3927	0.6459	0.0963	0.0028	0.0028	0.0106			
725	11	7	0.899	9.31	22.50	0.0010	0.0042	0.0042	0.1546	0.3322	0.03	0.05	0.14	6.11	0.1734	0.5887	
		6	-0.0349	0.0010	0.0042	0.0010	0.0042	0.0042	0.1546	0.3322	0.6384	0.0794	0.0029	0.0093	0.1583	0.5470	
		9	0.0329	0.0021	0.0042	0.0021	0.0042	0.0042	0.3322	0.6384	0.0924	0.0029	0.0029	0.0101			

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key →																				
725	12	7	0.898	9.31	33.79	0.1451	0.06	0.04	0.16	6.11	0.1609	0.6004	0.0527	0.0015	0.0067	0.3128	0.6388	0.0742	0.0023	0.0089	0.1783	0.5485	
		8	0.0464	0.0029	0.0057	0.3128	0.6156	0.0860	0.0030	0.0094													
725	13	7	0.899	9.30	44.99	0.1515	0.08	0.01	0.18	6.11	0.1476	0.5160	0.0557	0.0020	0.0084	0.2747	0.5585	0.0646	0.0019	0.0079	0.2044	0.5487	
		8	0.0557	0.0020	0.0066	0.2747	0.5585	0.0801	0.0032	0.0088													
725	14	7	0.900	9.30	56.29	0.1623	0.09	0.00	0.19	6.10	0.1469	0.6287	0.0743	0.0024	0.0089	0.1623	0.6030	0.0501	0.0014	0.0063	0.2298	0.5578	
		8	0.0680	0.0035	0.0077	0.2610	0.5699	0.0704	0.0032	0.0078													
725	15	7	0.901	9.30	67.49	0.1745	0.10	-0.02	0.20	6.09	0.1541	0.6160	0.0784	0.0027	0.0084	0.1745	0.5896	0.0323	0.0030	0.0068	0.2298	0.5578	
		8	0.0784	0.0035	0.0084	0.2412	0.5717	0.0594	0.0030	0.0068													
725	16	7	0.897	9.31	78.79	0.1922	0.11	-0.04	0.20	6.07	0.1485	0.5898	0.0788	0.0030	0.0089	0.1922	0.5866	0.0151	0.0004	0.0017	0.2522	0.6032	
		8	0.0788	0.0035	0.0089	0.2273	0.5679	0.0478	0.0024	0.0057													
725	17	7	0.898	9.31	89.99	0.1937	0.11	-0.06	0.20	6.05	0.7479	0.6375	0.0791	0.0030	0.0090	0.1937	0.5739	0.0002	0.0000	0.0002	0.2600	0.6375	
		8	0.0791	0.0035	0.0091	0.2223	0.5760	0.0318	0.0016	0.0040													
726	1	7	0.899	12.43	89.99	0.1574	0.13	-0.07	0.21	6.05	1.2355	0.6457	0.0969	0.0033	0.0108	0.1574	0.5608	0.0000	0.0000	0.0001	0.2654	0.6457	
		8	0.0969	0.0033	0.0108	0.1727	0.5608	0.0314	0.0016	0.0040													
726	2	7	0.901	12.43	78.78	0.1640	0.13	-0.04	0.21	6.08	0.1402	0.5678	0.0989	0.0032	0.0113	0.1640	0.5759	0.0205	0.0026	0.0059	0.2639	0.5957	
		8	0.0989	0.0032	0.0113	0.1683	0.5401	0.0502	0.0026	0.0059													
726	3	7	0.901	12.43	67.48	0.1596	0.13	-0.01	0.20	6.09	0.1403	0.6166	0.0985	0.0033	0.0096	0.1596	0.5739	0.0438	0.0031	0.0073	0.2388	0.5569	
		8	0.0985	0.0033	0.0096	0.1873	0.5439	0.0658	0.0031	0.0073													
726	4	7	0.900	12.43	56.29	0.1481	0.12	0.02	0.21	6.11	0.1307	0.6328	0.0944	0.0027	0.0088	0.1481	0.5906	0.0664	0.0017	0.0085	0.2045	0.5458	
		8	0.0944	0.0027	0.0088	0.2234	0.5457	0.0784	0.0032	0.0085													
726	5	7	0.898	12.43	44.99	0.1406	0.11	0.05	0.20	6.10	0.1316	0.6177	0.0855	0.0024	0.0080	0.1406	0.6181	0.0849	0.0022	0.0096	0.1619	0.5438	
		8	0.0855	0.0024	0.0080	0.2526	0.5469	0.0890	0.0022	0.0096													

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																			
726	6	7	0.900	12.44	33.80	0.1249	0.08	0.06	0.18	6.11	0.1400	0.5958	0.0701	0.0017	0.0089	0.6406	0.0955	0.0026	0.0113	0.1391	0.5469	
		8	0.0570	0.0035	0.0067	0.3129	0.5896	0.0949	0.0026	0.0103												
		9	0.0459	0.0013	0.0059	0.1455	0.6429	0.0991	0.0030	0.0115												
726	7	7	0.901	12.44	22.51	0.3112	0.05	0.07	0.15	6.11	0.1515	0.5826	0.0422	0.0026	0.0049	0.5883	0.0992	0.0025	0.0109	0.1291	0.5539	
		8	0.0232	0.0006	0.0028	0.1407	0.6753	0.1019	0.0031	0.0116												
		9	0.0207	0.0015	0.0026	0.3740	0.6418	0.1050	0.0026	0.0115												
726	8	7	0.898	12.44	11.31	0.1407	0.02	0.07	0.11	6.11	0.1530	0.5710	0.898	0.0006	0.0026	0.6753	0.0991	0.0026	0.0116	0.1251	0.5485	
		8	0.0232	0.0006	0.0028	0.3740	0.6418	0.1050	0.0026	0.0115												
		9	0.0207	0.0015	0.0026	0.3740	0.6418	0.1050	0.0026	0.0115												
726	9	7	0.898	12.45	0.00	-0.0964	-0.01	0.07	0.06	6.11	0.1551	0.5641	0.898	0.0002	0.0000	0.7719	0.0992	0.0030	0.0111	0.1202	0.5492	
		8	0.0223	-0.0000	0.0003	-0.0964	0.7719	0.0992	0.0030	0.0111												
		9	0.0001	0.0002	0.0000	12.6904	3.0168	0.1083	0.0026	0.0118												
726	10	7	0.898	12.45	-11.31	0.2225	0.05	0.07	0.02	6.11	0.1588	0.5601	0.898	0.0008	-0.0021	0.5774	0.0951	0.0030	0.0106	0.1130	0.5536	
		8	-0.0184	-0.0008	-0.0021	0.2225	0.5774	0.0951	0.0030	0.0123												
		9	-0.0188	-0.0010	-0.0025	0.2893	0.6793	0.1114	0.0025	0.0123												
726	11	7	0.899	12.45	-22.51	0.1907	0.08	0.06	-0.02	6.12	0.1736	0.5567	0.899	0.0015	-0.0052	0.6321	0.0893	0.0031	0.0099	0.1153	0.5571	
		8	-0.0414	-0.0015	-0.0052	0.1907	0.6321	0.0893	0.0031	0.0127												
		9	-0.0400	-0.0022	-0.0050	0.2858	0.6280	0.1148	0.0026	0.0127												
726	12	7	0.898	12.45	-33.80	0.1568	0.12	0.06	-0.04	6.12	0.1969	0.5532	0.898	0.0020	-0.0083	0.8375	0.0822	0.0032	0.0091	0.1257	0.5596	
		8	-0.0658	-0.0020	-0.0083	0.1568	0.8375	0.0822	0.0032	0.0129												
		9	-0.0554	-0.0031	-0.0063	0.2880	0.5705	0.1152	0.0028	0.0129												
726	13	7	0.897	12.45	-44.99	0.1473	0.14	0.05	-0.06	6.12	0.2245	0.5628	0.897	0.0025	-0.0105	0.6211	0.0741	0.0033	0.0085	0.1257	0.5628	
		8	-0.0849	-0.0025	-0.0105	0.1473	0.6211	0.0741	0.0033	0.0127												
		9	-0.0697	-0.0034	-0.0079	0.2501	0.5704	0.1110	0.0027	0.0127												
726	14	7	0.897	12.45	-56.29	0.1528	0.16	0.03	-0.07	6.10	0.2605	0.5745	0.897	0.0029	-0.0116	0.5996	0.0600	0.0031	0.0068	0.1111	0.5920	
		8	-0.0974	-0.0029	-0.0116	0.1528	0.5996	0.0600	0.0031	0.0119												
		9	-0.0795	-0.0033	-0.0089	0.2116	0.5624	0.1010	0.0022	0.0119												
726	15	7	0.898	12.46	-67.49	0.1589	0.17	0.01	-0.07	6.08	0.2717	0.6145	0.898	0.0032	-0.0119	0.5782	0.0429	0.0023	0.0052	0.0895	0.6218	
		8	-0.1029	-0.0032	-0.0119	0.1589	0.5782	0.0429	0.0023	0.0108												
		9	-0.0866	-0.0031	-0.0097	0.1803	0.5639	0.0875	0.0015	0.0108												
726	16	7	0.899	12.46	-78.79	0.1706	0.18	0.03	-0.07	6.06	0.2751	0.6318	0.899	0.0035	-0.0118	0.5743	0.0220	0.0012	0.0027	0.0829	0.6355	
		8	-0.1031	-0.0035	-0.0118	0.1706	0.5743	0.0220	0.0012	0.0084												
		9	-0.0929	-0.0028	-0.0102	0.1540	0.5485	0.0665	0.0011	0.0084												

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																
726	17	0.900	12.46	-90.00	0.1798	-0.17	-0.06	-0.07	6.04	0.1629	0.1759							
	8	-0.0983	-0.0035	-0.0113	0.1358	0.5774	0.0022	0.0000	0.0000	0.0755	0.6285							
	9	-0.0985	-0.0026	-0.0102		0.5206	0.0426	0.0006	0.0053									
727	6	0.898	12.44	-90.01	0.1615	-0.17	-0.06	-0.09	9.04	0.3210	0.2490							
	8	-0.1024	-0.0033	-0.0114	0.1588	0.5592	0.0022	0.0001	0.0001	0.0720	0.6536							
	9	-0.0940	-0.0029	-0.0105		0.5625	0.0643	0.0009	0.0084									
727	7	0.897	12.45	-78.79	0.1658	-0.17	-0.02	-0.08	9.06	0.2885	0.6402							
	8	-0.1044	-0.0034	-0.0119	0.1623	0.5706	0.0221	0.0012	0.0028	0.0858	0.6281							
	9	-0.0909	-0.0029	-0.0100		0.5526	0.0848	0.0014	0.0106									
727	8	0.897	12.45	-67.49	0.1565	-0.17	0.01	-0.08	9.08	0.2773	0.6159							
	8	-0.1036	-0.0032	-0.0120	0.1807	0.5798	0.0429	0.0023	0.0052	0.0988	0.6040							
	9	-0.0867	-0.0031	-0.0095		0.5514	0.1003	0.0019	0.0121									
727	9	0.896	12.45	-56.29	0.1467	-0.15	0.04	-0.08	9.11	0.2661	0.5792							
	8	-0.0978	-0.0028	-0.0116	0.2196	0.5956	0.0599	0.0031	0.0069	0.1231	0.5811							
	9	-0.0783	-0.0034	-0.0088		0.5626	0.1113	0.0027	0.0129									
727	10	0.896	12.44	-44.99	0.1449	-0.14	0.05	-0.07	9.11	0.2288	0.5661							
	8	-0.0843	-0.0024	-0.0104	0.2523	0.6207	0.0738	0.0033	0.0083	0.1218	0.5662							
	9	-0.0687	-0.0034	-0.0076		0.5590	0.1174	0.0028	0.0133									
727	11	0.896	12.44	-33.80	0.1540	-0.11	0.06	-0.05	9.10	0.2006	0.5571							
	8	-0.0648	-0.0019	-0.0081	0.2928	0.6312	0.0819	0.0032	0.0091	0.1097	0.5571							
	9	-0.0544	-0.0031	-0.0061		0.5658	0.1190	0.0026	0.0133									
727	12	0.897	12.44	-22.51	0.1884	-0.08	0.06	-0.02	9.10	0.1765	0.5588							
	8	-0.0400	-0.0015	-0.0049	0.2886	0.6217	0.0892	0.0031	0.0099	0.1085	0.5592							
	9	-0.0392	-0.0022	-0.0048		0.6173	0.1140	0.0024	0.0126									
727	13	0.897	12.44	-11.30	0.2163	-0.05	0.06	0.02	9.10	0.1617	0.5616							
	8	-0.0174	-0.0007	-0.0019	0.2972	0.5511	0.0949	0.0030	0.0106	0.1177	0.5616							
	9	-0.0175	-0.0010	-0.0023		0.6673	0.1135	0.0026	0.0124									
727	14	0.898	12.44	0.00	-0.0219	-0.02	0.07	0.06	9.10	0.1565	0.5636							
	8	0.0032	-0.0000	0.0004	0.7601	0.7414	0.0990	0.0031	0.0111	0.1167	0.5636							
	9	0.0018	0.0002	0.0002		0.5573	0.1127	0.0026	0.0124									
727	15	0.897	12.44	11.31	0.1603	0.02	0.08	0.10	9.10	0.1565	0.5724							
	8	0.0235	0.0007	0.0030	0.3420	0.6396	0.1013	0.0031	0.0116	0.1203	0.5724							
	9	0.0221	0.0015	0.0027		0.6218	0.1102	0.0026	0.0121									

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key													
727 16	7	0.097	12.42	22.51	0.1473	0.05	0.07	0.14	9.10	0.1532	0.5811				
	8	0.0471	0.0013	0.0061	0.3231	0.6567	0.0995	0.0030	0.0115	0.1252	0.5470				
	9	0.0422	0.0027	0.0052	0.0052	0.6244	0.1061	0.0026	0.0116						
727 17	7	0.099	12.43	33.00	0.1335	0.08	0.05	0.16	9.09	0.1422	0.5941				
	8	0.0705	0.0018	0.0091	0.2925	0.6510	0.0950	0.0027	0.0112	0.1342	0.5490				
	9	0.0593	0.0034	0.0067	0.0067	0.5670	0.0992	0.0026	0.0109						
727 18	7	0.097	12.42	44.99	0.1408	0.10	0.05	0.19	9.09	0.1328	0.6143				
	8	0.0857	0.0024	0.0105	0.2550	0.6159	0.0846	0.0022	0.0103	0.1404	0.5527				
	9	0.0731	0.0037	0.0081	0.0081	0.5599	0.0921	0.0025	0.0101						
727 19	7	0.098	12.42	56.29	0.1537	0.12	0.02	0.19	9.09	0.1326	0.6259				
	8	0.0932	0.0028	0.0110	0.2190	0.5944	0.0664	0.0017	0.0083	0.1679	0.5486				
	9	0.0812	0.0035	0.0089	0.0089	0.5479	0.0860	0.0028	0.0094						
727 20	7	0.098	12.41	67.48	0.1654	0.13	-0.00	0.20	9.09	0.1470	0.6163				
	8	0.0973	0.0032	0.0113	0.1855	0.5805	0.0434	0.0012	0.0053	0.2189	0.5487				
	9	0.0887	0.0032	0.0097	0.0097	0.5506	0.0754	0.0033	0.0082						
727 21	7	0.097	12.42	78.78	0.1563	0.13	-0.03	0.21	9.07	0.1561	0.5635				
	8	0.0990	0.0030	0.0111	0.1798	0.5641	0.0201	0.0006	0.0022	0.2490	0.5664				
	9	0.0921	0.0033	0.0106	0.0106	0.5754	0.0615	0.0030	0.0069						
727 22	7	0.096	12.42	89.99	0.1610	0.12	-0.07	0.21	9.06	-0.7520	4.7600				
	8	0.0961	0.0030	0.0108	0.1687	0.5618	-0.0002	0.0000	-0.0002	0.2696	0.6220				
	9	0.0970	0.0032	0.0108	0.0108	0.5601	0.0456	0.0024	0.0056						
72A 1	7	0.098	9.30	89.99	0.2008	0.12	-0.05	0.19	9.06	-1.6980	23.3068				
	8	0.0777	0.0031	0.0090	0.2182	0.5792	-0.0000	0.0000	-0.0002	0.2573	0.6174				
	9	0.0800	0.0034	0.0091	0.0091	0.5735	0.0474	0.0024	0.0058						
72B 2	7	0.098	9.30	78.79	0.1987	0.12	-0.04	0.19	9.08	0.1574	0.5966				
	8	0.0777	0.0030	0.0091	0.2222	0.5899	0.0152	0.0004	0.0018	0.2635	0.5792				
	9	0.0790	0.0035	0.0088	0.0088	0.5619	0.0584	0.0030	0.0067						
72A 3	7	0.097	9.30	67.49	0.1839	0.11	-0.01	0.19	9.09	0.1563	0.6177				
	8	0.0777	0.0028	0.0092	0.2327	0.5961	0.0329	0.0010	0.0040	0.2265	0.5564				
	9	0.0757	0.0035	0.0084	0.0084	0.5593	0.0706	0.0032	0.0078						
72A 4	7	0.097	9.30	56.29	0.1651	0.10	0.00	0.18	9.09	0.1515	0.6315				
	8	0.0732	0.0024	0.0088	0.2654	0.5996	0.0504	0.0015	0.0063	0.2018	0.5488				
	9	0.0676	0.0036	0.0078	0.0078	0.5815	0.0799	0.0032	0.0087						

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key →														
728	5	7	0.898	9.30	44.99	0.1548	0.301	0.09	0.02	0.17	9.09	0.1510	0.6170			
		8	0.0667	0.0020	0.0084	0.2827	0.6301	0.0645	0.0019	0.0079		0.1713	0.5497			
		9	0.0593	0.0033	0.0068		0.5791	0.0868	0.0029	0.0095						
728	6	7	0.897	9.30	33.79	0.1452	0.337	0.04	0.15	5.08	0.1679	0.6035				
		8	0.0533	0.0015	0.0067	0.3047	0.6337	0.0736	0.0024	0.0088		0.1490	0.5510			
		9	0.0477	0.0029	0.0057		0.6055	0.0927	0.0027	0.0102						
728	7	7	0.897	9.30	22.50	0.1586	0.374	0.05	0.12	9.08	0.1777	0.5899				
		8	0.0359	0.0011	0.0045	0.3020	0.6374	0.0790	0.0028	0.0093		0.1356	0.5506			
		9	0.0350	0.0021	0.0042		0.5995	0.0964	0.0026	0.0106						
728	8	7	0.895	9.31	11.30	0.1757	0.212	0.05	0.09	9.09	0.1883	0.5818				
		8	0.0183	0.0006	0.0022	0.3811	0.6727	0.0815	0.0030	0.0094		0.1281	0.5518			
		9	0.0169	0.0012	0.0022		0.6727	0.0996	0.0025	0.0109						
728	9	7	0.896	9.32	-0.00	-0.0801	-0.00	0.06	0.06	9.09	0.1930	0.5770				
		8	0.0033	-0.0000	0.0003	1.1524	0.822	0.0815	0.0031	0.0094		0.1264	0.5551			
		9	0.0014	0.0003	0.0003		1.0243	0.1014	0.0025	0.0112						
728	10	7	0.897	9.32	-11.30	0.3138	0.5992	0.05	0.02	9.09	0.2009	0.5753				
		8	-0.0115	-0.0007	-0.0013	0.2132	0.5992	0.0793	0.0031	0.0091		0.1241	0.5549			
		9	-0.0134	-0.0005	-0.0014		0.5500	0.1040	0.0025	0.0115						
728	11	7	0.896	9.31	-22.50	0.2209	0.6197	0.05	-0.00	9.10	0.2145	0.5745				
		8	-0.0297	-0.0013	-0.0036	0.2588	0.5812	0.0752	0.0032	0.0086		0.1279	0.5546			
		9	-0.0295	-0.0015	-0.0034		0.5812	0.1058	0.0027	0.0117						
728	12	7	0.897	9.31	-33.79	0.1776	0.6323	0.05	-0.03	9.10	0.2323	0.5745				
		8	-0.0489	-0.0017	-0.0061	0.3029	0.6323	0.0685	0.0031	0.0078		0.1369	0.5574			
		9	-0.0424	-0.0025	-0.0054		0.6444	0.1057	0.0028	0.0117						
728	13	7	0.896	9.31	-44.99	0.1778	0.6246	0.03	-0.06	9.10	0.2471	0.5882				
		8	-0.0632	-0.0022	-0.0079	0.2826	0.6246	0.0607	0.0030	0.0071		0.1385	0.5641			
		9	-0.0547	-0.0030	-0.0062		0.5714	0.1035	0.0028	0.0116						
728	14	7	0.898	9.32	-56.29	0.1765	0.6084	0.02	-0.06	9.09	0.2618	0.6017				
		8	-0.0742	-0.0026	-0.0090	0.2653	0.6084	0.0491	0.0025	0.0059		0.1397	0.5727			
		9	-0.0638	-0.0033	-0.0072		0.5681	0.0969	0.0027	0.0111						
728	15	7	0.897	9.32	-67.49	0.1867	0.5964	-0.00	-0.08	9.07	0.2705	0.6254				
		8	-0.0809	-0.0030	-0.0096	0.2404	0.5964	0.0350	0.0018	0.0043		0.1196	0.5915			
		9	-0.0721	-0.0034	-0.0081		0.5634	0.0891	0.0021	0.0105						

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																		
728	16	0.898	9.32	-78.79	0.1953	-0.15	-0.03	-0.08	9.06	0.2696	0.6258	-0.837	-0.0032	-0.0097	0.5850	0.0185	0.0010	0.0023	0.1055	0.6128
	8	-0.0784	-0.0033	-0.0086	0.2153	0.5527	0.0772	0.0016	0.0094											
728	17	0.897	9.32	-89.99	0.2002	-0.15	-0.06	-0.08	9.04	0.2543	0.3800	-0.831	-0.0033	-0.0095	0.5754	0.0033	0.0001	0.0002	0.0947	0.6370
	8	-0.0799	-0.0033	-0.0090	0.2106	0.5659	0.0633	0.0011	0.0080											
729	1	0.897	6.21	-69.99	0.2257	-0.12	-0.06	-0.05	9.05	0.2963	0.4432	-0.603	-0.0027	-0.0072	0.5982	0.0039	0.0002	0.0003	0.1334	0.6200
	8	-0.0583	-0.0029	-0.0071	0.2509	0.6157	0.0606	0.0016	0.0075											
729	2	0.898	6.21	-78.79	0.2323	-0.12	-0.03	-0.04	9.06	0.2715	0.6287	-0.584	-0.0027	-0.0072	0.6234	0.0148	0.0008	0.0018	0.1387	0.6030
	8	-0.0572	-0.0027	-0.0067	0.2418	0.5917	0.0698	0.0019	0.0084											
729	3	0.896	6.21	-67.49	0.2059	-0.11	-0.02	-0.05	9.07	0.2618	0.6298	-0.564	-0.0023	-0.0070	0.6216	0.0254	0.0013	0.0032	0.1479	0.5827
	8	-0.0521	-0.0027	-0.0063	0.2602	0.6086	0.0777	0.0023	0.0092											
729	4	0.900	6.21	-56.29	0.1864	-0.10	-0.00	-0.03	9.08	0.2597	0.6306	-0.451	-0.0019	-0.0055	0.6326	0.0359	0.0018	0.0045	0.1573	0.5793
	8	-0.0451	-0.0024	-0.0055	0.2674	0.6111	0.0838	0.0026	0.0097											
729	5	0.897	6.21	-44.99	0.1927	-0.09	0.01	-0.02	9.08	0.2473	0.6238	-0.426	-0.0016	-0.0045	0.6446	0.0454	0.0022	0.0056	0.1648	0.5656
	8	-0.0367	-0.0020	-0.0045	0.2751	0.6181	0.0877	0.0028	0.0099											
729	6	0.898	6.21	-33.79	0.1830	-0.07	0.01	-0.00	9.09	0.2398	0.6101	-0.327	-0.0011	-0.0041	0.6321	0.0517	0.0024	0.0063	0.1632	0.5611
	8	-0.0264	-0.0015	-0.0034	0.2918	0.6570	0.0913	0.0029	0.0102											
729	7	0.897	6.22	-22.49	0.2097	-0.05	0.02	0.02	9.09	0.2316	0.6033	-0.204	-0.0008	-0.0022	0.6346	0.0565	0.0026	0.0068	0.1592	0.5571
	8	-0.0164	-0.0009	-0.0022	0.2831	0.6834	0.0941	0.0029	0.0104											
729	8	0.897	6.22	-11.30	0.1948	-0.03	0.03	0.04	9.09	0.2243	0.6036	-0.092	-0.0003	-0.0010	0.5625	0.0588	0.0026	0.0071	0.1613	0.5501
	8	-0.0092	-0.0002	-0.0008	0.1922	0.6954	0.0941	0.0030	0.0103											
729	9	0.898	6.22	-0.00	-0.0825	-0.02	0.02	0.06	9.09	0.2158	0.6037	-0.011	-0.0000	-0.0001	0.5123	0.0596	0.0035	0.0072	0.1649	0.5512
	8	0.0016	-0.0004	0.0003	-1.2970	1.1923	0.0930	0.0030	0.0102											

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd																			
729	10	7	0.898	6.21	11.29	0.1638	0.00	0.02	0.08	9.09	0.2017	0.6050								
		8	0.0122	0.0004	0.0017	0.3454	0.6970	0.0590	0.0023	0.0071	0.1700	0.5527								
		9	0.0143	0.0009	0.0015		0.5425	0.0905	0.0030	0.0100										
729	11	7	0.898	6.21	22.49	0.1430	0.02	0.01	0.10	9.09	0.1832	0.6178								
		8	0.0243	0.0006	0.0030	0.3657	0.6220	0.0568	0.0020	0.0070	0.1788	0.5519								
		9	0.0238	0.0017	0.0033		0.6969	0.0877	0.0031	0.0096										
729	12	7	0.897	6.20	33.79	0.1649	0.04	0.00	0.12	9.10	0.1679	0.6327								
		8	0.0348	0.0011	0.0046	0.2853	0.6654	0.0519	0.0017	0.0065	0.1897	0.5529								
		9	0.0361	0.0020	0.0042		0.5867	0.0845	0.0032	0.0093										
729	13	7	0.896	6.20	44.99	0.1827	0.06	-0.00	0.13	9.09	0.1484	0.6404								
		8	0.0442	0.0016	0.0059	0.2850	0.6761	0.0447	0.0013	0.0057	0.2040	0.5557								
		9	0.0438	0.0025	0.0052		0.6009	0.0781	0.0031	0.0086										
729	14	7	0.897	6.20	56.29	0.1902	0.07	-0.01	0.15	9.09	0.1346	0.6316								
		8	0.0519	0.0019	0.0069	0.2692	0.6886	0.0343	0.0009	0.0043	0.2207	0.5639								
		9	0.0510	0.0027	0.0059		0.5784	0.0722	0.0031	0.0061										
729	15	7	0.897	6.21	67.49	0.1925	0.08	-0.03	0.16	9.07	0.1194	0.6258								
		8	0.0578	0.0022	0.0076	0.2652	0.6577	0.0227	0.0005	0.0028	0.2387	0.5711								
		9	0.0552	0.0029	0.0064		0.5844	0.0652	0.0031	0.0074										
729	16	7	0.898	6.20	78.79	0.2024	0.09	-0.05	0.16	9.07	0.2431	0.6058								
		8	0.0595	0.0024	0.0076	0.2462	0.6398	0.0091	0.0004	0.0011	0.2420	0.5902								
		9	0.0589	0.0029	0.0066		0.5624	0.0577	0.0027	0.0068										
729	17	7	0.898	6.21	89.99	0.2092	0.09	-0.07	0.16	9.05	-0.4034	-0.5954								
		8	0.0595	0.0024	0.0075	0.2712	0.6316	0.0008	-0.0000	-0.0001	0.7380	2.6306								
		9	0.0561	0.0030	0.0070		0.6260	0.0486	0.0023	0.0060	0.2205	0.6170								
730	1	7	0.897	3.08	89.99	0.2071	0.03	-0.06	0.11	9.05	0.5119	0.4433								
		8	0.0266	0.0011	0.0034	0.2817	0.6472	-0.0005	-0.0000	-0.0002	0.2210	0.6052								
		9	0.0312	0.0017	0.0040		0.6457	0.0516	0.0022	0.0063										
730	2	7	0.898	3.08	78.79	0.2129	0.04	-0.05	0.11	9.06	0.4112	0.5902								
		8	0.0267	0.0011	0.0035	0.2908	0.6613	0.0024	0.0002	0.0002	0.2221	0.5970								
		9	0.0311	0.0018	0.0040		0.6535	0.0562	0.0024	0.0068										
730	3	7	0.898	3.07	67.49	0.1954	0.04	-0.05	0.11	9.07	0.4112	0.5902								
		8	0.0266	0.0010	0.0034	0.2854	0.6509	0.0066	0.0005	0.0007	0.2221	0.5970								
		9	0.0302	0.0017	0.0037		0.6232	0.0603	0.0026	0.0072										

See Key

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key													
730	4	7	8	9	56.29	0.1933	0.6274	0.03	-0.04	0.11	9.07	0.1642	0.6533		
					0.0033	0.3080	0.6397	0.0027	0.0146	0.0005	0.0019	0.2165	0.5630		
					0.0034			0.0644		0.0027	0.0075				
730	5	7	8	9	44.99	0.1471	0.6206	0.02	-0.03	0.10	9.07	0.1741	0.6485		
					0.0028	0.3147	0.6135	0.0028	0.0203	0.0007	0.0026	0.2154	0.5615		
					0.0028			0.0672		0.0028	0.0078				
730	6	7	8	9	33.79	0.1289	0.6286	0.01	-0.02	0.09	9.07	0.1852	0.6517		
					0.0023	0.3310	0.5779	0.0029	0.0255	0.0009	0.0033	0.2087	0.5729		
					0.0021			0.0705		0.0029	0.0080				
730	7	7	8	9	22.49	0.1196	0.6684	0.00	-0.01	0.08	9.07	0.1954	0.6480		
					0.0017	0.3950	0.6283	0.0019	0.0293	0.0019	0.0038	0.2046	0.5701		
					0.0014			0.0728		0.0019	0.0083				
730	8	7	8	9	11.29	0.1191	0.6467	0.00	-0.01	0.07	9.08	0.2063	0.6427		
					0.0008	0.5441	0.7247	0.0030	0.0320	0.0013	0.0041	0.2031	0.5713		
					0.0009			0.0743		0.0030	0.0084				
730	9	7	8	9	-0.00	-0.2305	-0.3809	-0.01	-0.01	0.06	9.08	0.2196	0.6411		
					0.0001	0.4320	0.0455	0.0014	0.0334	0.0030	0.0042	0.2001	0.5702		
					0.0000			0.0752		0.0030	0.0085				
730	10	7	8	9	-11.30	0.3400	0.9088	-0.02	-0.01	0.05	9.08	0.2277	0.6407		
					0.0006	0.6683	3.0789	0.002	0.0331	0.0029	0.0042	0.1947	0.5698		
					-0.0006			0.0789		0.0029	0.0086				
730	11	7	8	9	-22.50	0.1917	0.6426	-0.03	-0.01	0.03	9.08	0.2361	0.6381		
					0.0012	0.3113	0.9551	0.0026	0.0315	0.0014	0.0040	0.1923	0.5740		
					-0.0012			0.9551		0.0029	0.0087				
730	12	7	8	9	-33.79	0.1752	0.6278	-0.04	-0.01	0.02	9.08	0.2440	0.6389		
					0.0019	0.2625	0.7415	0.004	0.0284	0.0028	0.0036	0.1600	0.5792		
					-0.0017			0.7415		0.0028	0.0087				
730	13	7	8	9	-44.99	0.2038	0.6513	-0.05	-0.02	0.01	9.07	0.2511	0.6349		
					0.0026	0.2139	0.6011	0.005	0.0244	0.0012	0.0031	0.1852	0.5844		
					-0.0021			0.6011		0.0012	0.0086				
730	14	7	8	9	-56.29	0.2157	0.6587	-0.06	-0.03	0.01	9.07	0.2507	0.6364		
					0.0033	0.2229	0.5875	0.006	0.0196	0.0009	0.0025	0.1638	0.5919		
					-0.0025			0.5875		0.0026	0.0084				

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key															
730	15	7	0.898	3.09	-67.49	0.2452	-0.07	-0.04	0.00	9.06	0.2514	0.6322						
	8	8	-0.0266	-0.0014	-0.0039	0.2621	0.645	0.0142	0.0007	0.0018	0.1774	0.5933						
	9	9	-0.0230	-0.0012	-0.0029	0.2621	0.6506	0.0682	0.0024	0.0080								
730	16	7	0.898	3.10	-78.79	0.2394	-0.08	-0.05	0.00	9.06	0.3090	0.5732						
	8	8	-0.0322	-0.0015	-0.0042	0.2435	0.636	0.080	0.0004	0.0009	0.1721	0.5963						
	9	9	-0.0261	-0.0012	-0.0031	0.2435	0.6041	0.0645	0.0022	0.0077								
730	17	7	0.900	3.10	-89.99	0.2432	-0.08	-0.05	0.00	9.05	0.3449	0.3799						
	8	8	-0.0336	-0.0016	-0.0043	0.2657	0.492	0.031	0.0002	0.0002	0.1681	0.6077						
	9	9	-0.0267	-0.0014	-0.0034	0.2657	0.6473	0.0597	0.0020	0.0072								
731	1	7	0.898	-0.00	-90.00	0.5208	-0.03	-0.05	0.05	9.06	0.3298	0.3503						
	8	8	-0.0032	-0.0003	-0.0005	-1.3486	0.292	0.033	0.0002	0.0002	0.1948	0.6058						
	9	9	-0.0011	0.0003	0.0001	-1.3486	-0.5986	0.0584	0.0022	0.0070								
731	2	7	0.900	-0.00	-78.80	0.2015	-0.03	-0.05	0.05	9.05	0.3615	0.4130						
	8	8	-0.0041	-0.0001	-0.0004	0.1398	0.222	0.035	0.0002	0.0002	0.1968	0.6081						
	9	9	-0.0024	0.0000	-0.0002	0.1398	-0.4413	0.0581	0.0022	0.0070								
731	3	7	0.900	0.00	-67.50	0.3663	-0.02	-0.05	0.05	9.06	0.3599	0.4313						
	8	8	-0.0031	-0.0002	-0.0004	-3.0553	0.685	0.037	0.0002	0.0003	0.1985	0.6134						
	9	9	-0.0003	0.0002	0.0000	-3.0553	-0.2045	0.0575	0.0022	0.0070								
731	4	7	0.898	0.00	-56.30	0.4261	-0.03	-0.05	0.05	9.05	0.3744	0.4554						
	8	8	-0.0024	-0.0002	-0.0003	0.8875	0.120	0.037	0.0002	0.0003	0.1997	0.6128						
	9	9	-0.0008	0.0001	-0.0000	0.8875	-0.4384	0.0570	0.0022	0.0069								
731	5	7	0.899	0.00	-45.00	0.5404	-0.02	-0.05	0.05	9.06	0.4097	0.4677						
	8	8	-0.0015	-0.0001	-0.0001	3.4099	0.507	0.035	0.0002	0.0003	0.2028	0.6185						
	9	9	-0.0002	0.0001	0.0000	3.4099	0.1090	0.0560	0.0022	0.0069								
731	6	7	0.900	0.01	-33.80	4.4504	-0.02	-0.05	0.05	9.06	0.4310	0.4776						
	8	8	-0.0002	-0.0002	-0.0000	0.9355	1.2961	0.033	0.0002	0.0003	0.2018	0.6210						
	9	9	-0.0008	0.0001	-0.0000	0.9355	-0.1544	0.0550	0.0022	0.0068								
731	7	7	0.900	0.01	-22.50	-1.3953	-0.02	-0.06	0.05	9.05	0.4901	0.5354						
	8	8	0.0003	-0.0000	0.0002	1.6617	3.382	0.031	0.0003	0.0003	0.2016	0.6201						
	9	9	0.0006	-0.0002	0.0000	1.6617	0.3603	0.0547	0.0022	0.0067								
731	8	7	0.901	0.02	-11.30	-0.6770	-0.02	-0.06	0.05	9.05	0.4527	0.5135						
	8	8	0.0016	-0.0002	0.0001	1.5152	0.589	0.031	0.0002	0.0003	0.2016	0.6225						
	9	9	0.0008	-0.0002	0.0000	1.5152	0.4048	0.0543	0.0021	0.0067								

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

See Key →

Run	Pt.	Cd																								
731	9	7	0.899	-0.0002	0.02	-0.001	-0.5634	0.3466	-0.06	0.05	9.05	0.4553	0.5004													
		8	0.0025	0.0003	0.0001	0.0001	9.1330	0.3466	0.0030	0.0021	0.0067	0.1988	0.6203													
		9	0.0001	0.0003	0.0001	0.0001		5.6721	0.0547	0.0021	0.0067															
731	10	7	0.899	-0.0003	0.02	11.29	-0.5686	0.8654	-0.05	0.05	9.05	0.4913	0.5113													
		8	0.0014	0.0003	0.0002	0.0002	-2.8741	0.8654	0.0026	0.0021	0.0067	0.2006	0.6204													
		9	0.0005	0.0003	0.0002	0.0002		1.8995	0.0545	0.0021	0.0067															
731	11	7	0.897	-0.0003	0.01	22.49	-0.8180	0.412	-0.06	0.05	9.05	0.4993	0.4418													
		8	0.0006	0.0003	0.0002	0.0002	-1.8401	2.0431	0.0022	0.0021	0.0067	0.2005	0.6204													
		9	0.0009	0.0003	0.0002	0.0002		1.2451	0.0546	0.0021	0.0067															
731	12	7	0.898	-0.0004	0.01	33.79	-0.2766	0.01	-0.07	0.05	9.05	0.5746	0.3112													
		8	0.0004	0.0004	0.0001	0.0001	1.2766	0.0454	0.0015	0.0001	0.0067	0.2006	0.6204													
		9	0.0000	0.0004	0.0003	0.0003	-0.2836	0.2620	0.0546	0.0021	0.0067															
731	13	7	0.900	-0.0002	0.00	44.99	1.0812	0.0379	-0.06	0.05	9.05	0.5123	0.0189													
		8	0.0010	0.0002	0.0000	0.0000	2.3238	0.0490	0.0010	0.0022	0.0067	0.2027	0.6205													
		9	0.0007	0.0003	0.0002	0.0002		1.4907	0.0543	0.0022	0.0067															
731	14	7	0.898	-0.0002	0.00	56.29	0.5109	0.2751	-0.07	0.06	9.05	4.3725	1.3289													
		8	0.0020	0.0002	0.0001	0.0001	1.3882	0.7169	0.0000	0.0021	0.0067	0.1993	0.6176													
		9	0.0011	0.0003	0.0001	0.0001		0.9598	0.0545	0.0021	0.0067															
731	15	7	0.899	-0.0002	0.00	67.49	0.6095	0.4354	-0.06	0.05	9.05	-0.0320	2.2428													
		8	0.0021	0.0002	0.0001	0.0001	1.7906	0.9398	0.0005	0.0021	0.0067	0.1990	0.6185													
		9	0.0009	0.0003	0.0001	0.0001		0.9584	0.0543	0.0021	0.0067															
731	16	7	0.899	-0.0001	0.00	78.79	0.2217	0.2447	-0.06	0.05	9.05	0.1249	1.5983													
		8	0.0032	0.0001	0.0001	0.0001	1.8437	0.9584	0.0009	0.0021	0.0067	0.1991	0.6200													
		9	0.0008	0.0003	0.0001	0.0001		0.9584	0.0541	0.0021	0.0067															
731	17	7	0.899	-0.0002	0.00	89.99	0.4078	0.2461	-0.06	0.05	9.05	0.2278	1.9111													
		8	0.0026	0.0002	0.0001	0.0001	2.2687	1.1348	0.0006	0.0021	0.0066	0.1950	0.6156													
		9	0.0006	0.0003	0.0001	0.0001		0.9584	0.0541	0.0021	0.0066															
732	1	7	0.899	-0.0014	0.11	90.00	0.2416	0.6255	-0.06	0.00	9.05	0.1932	-0.7294													
		8	0.0309	0.0014	0.0038	0.0038	0.2321	0.5920	0.0009	0.0000	0.0001	0.1671	0.6206													
		9	0.0275	0.0012	0.0032	0.0032		0.5920	0.0565	0.0018	0.0078															
732	2	7	0.898	-0.0013	0.11	78.80	0.2187	0.5875	-0.07	0.00	9.07	0.2874	1.0545													
		8	0.0310	0.0013	0.0036	0.0036	0.2349	0.6063	0.0035	0.0017	0.0067	0.1621	0.6389													
		9	0.0266	0.0012	0.0032	0.0032		0.6063	0.0525	0.0017	0.0067															

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																
732	3	0.900	-3.10	67.50	0.2271	-0.06	-0.08	0.0003	0.01	9.03	0.2080	0.7746	0.0246	-0.0011	-0.0030	-0.0014	0.1571	0.6543
	8	0.899	-3.10	56.30	0.2506	-0.06	-0.09	0.0005	0.00	9.03	0.2109	0.7237	0.0254	-0.0012	-0.0030	-0.0019	0.1594	0.6653
	9	0.899	-3.10	45.00	0.2566	-0.06	-0.09	0.0007	0.02	9.03	0.2228	0.6865	0.0216	-0.0009	-0.0030	-0.0057	0.1681	0.6680
732	5	0.899	-3.09	33.80	0.3087	-0.05	-0.09	0.0009	0.03	9.03	0.2194	0.6682	0.0214	-0.0004	-0.0015	-0.0028	0.1793	0.6736
	8	0.899	-3.08	22.49	0.3053	-0.03	-0.11	0.0010	0.03	9.02	0.2205	0.6639	0.0157	-0.0004	-0.0012	-0.0044	0.1909	0.6789
	9	0.899	-3.08	11.29	0.3532	-0.03	-0.11	0.0011	0.04	9.01	0.2143	0.6597	0.0140	-0.0003	-0.0004	-0.0037	0.2000	0.6786
732	7	0.900	-3.08	0.000	-8.8591	3.8670	-0.11	0.05	0.05	9.01	0.2068	0.6615	0.0107	-0.0002	-0.0011	-0.0036	0.2075	0.6838
	8	0.899	-3.08	0.000	0.0530	0.7170	-0.10	0.06	0.06	9.02	0.1959	0.6577	0.0045	0.0004	-0.0011	-0.0037	0.2140	0.6775
	9	0.899	-3.07	-22.50	0.1931	0.6845	-0.10	0.07	0.07	9.02	0.1780	0.6625	0.0050	0.0007	-0.0012	-0.0039	0.2225	0.6785
732	10	0.899	-3.08	0.000	0.1923	0.6335	-0.09	0.08	0.08	9.03	0.1530	0.6667	0.0096	0.0005	-0.0014	-0.0042	0.2315	0.6768
	8	0.899	-3.08	0.000	0.3311	0.6233	-0.09	0.08	0.08	9.03	0.1390	0.6986	0.0100	0.0009	-0.0014	-0.0042	0.2312	0.6600
	9	0.899	-3.08	-45.00	0.2066	0.6217	-0.09	0.08	0.08	9.03	0.1390	0.6986	0.0100	0.0007	-0.0011	-0.0023	0.2312	0.6600

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key											
732	14	7	0.898	-3.08	-56.30	0.2104	0.01	-0.0123	0.09	9.04	0.1133	0.7234	
		8	0.0221	0.0009	0.0027	0.2701	0.6292	-0.0401	-0.002	-0.0017	0.2305	0.6531	
		9	0.0238	0.0012	0.0028		0.6030		0.0018	0.0052			
732	15	7	0.897	-3.09	-67.50	0.2153	0.02	-0.07	0.10	9.05	0.2037	0.7628	
		8	0.0247	0.0010	0.0031	0.2543	0.6359	-0.0053	-0.002	-0.0008	0.2255	0.6321	
		9	0.0274	0.0013	0.0032		0.5956	0.0456	0.0020	0.0057			
732	16	7	0.899	-3.09	-78.80	0.1849	0.02	-0.07	0.11	9.05	0.2272	1.9183	
		8	0.0267	0.0009	0.0032	0.2768	0.6097	-0.0007	-0.000	-0.0002	0.2241	0.6234	
		9	0.0278	0.0015	0.0035		0.6410	0.0504	0.0022	0.0062			
732	17	7	0.898	-3.09	-90.00	0.1815	0.01	-0.06	0.10	9.06	0.5828	0.3498	
		8	0.0272	0.0009	0.0033	0.2723	0.6133	0.0022	0.002	0.0001	0.2188	0.6065	
		9	0.0285	0.0015	0.0035		0.6188	0.0555	0.0024	0.0067			
733	1	7	0.899	-3.09	-90.00	0.1952	0.01	-0.06	0.10	15.10	0.5247	0.3770	
		8	0.0268	0.0010	0.0033	0.2743	0.6286	0.0027	0.002	0.0002	0.2004	0.5627	
		9	0.0305	0.0016	0.0038		0.6280	0.0784	0.0031	0.0088			
733	2	7	0.899	-3.08	-78.80	0.1999	0.01	-0.06	0.11	15.09	0.0861	2.1398	
		8	0.0261	0.0010	0.0033	0.2633	0.6323	-0.0005	-0.000	-0.0002	0.2073	0.5696	
		9	0.0307	0.0016	0.0037		0.6147	0.0758	0.0031	0.0086			
733	3	7	0.899	-3.08	-67.50	0.1825	0.00	-0.08	0.10	15.09	0.2355	0.7890	
		8	0.0251	0.0009	0.0030	0.2552	0.5978	-0.0046	-0.002	-0.0007	0.2158	0.5737	
		9	0.0296	0.0015	0.0036		0.6081	0.0718	0.0031	0.0082			
733	4	7	0.899	-3.06	-56.30	0.2050	-0.00	-0.08	0.10	15.09	0.1100	0.7262	
		8	0.0218	0.0008	0.0026	0.2470	0.6183	-0.0116	-0.002	-0.0016	0.2216	0.5807	
		9	0.0269	0.0013	0.0031		0.5916	0.0682	0.0030	0.0079			
733	5	7	0.900	-3.06	-45.00	0.2073	-0.00	-0.09	0.09	15.09	0.1309	0.6954	
		8	0.0180	0.0007	0.0022	0.2640	0.6295	-0.0166	-0.004	-0.0023	0.2253	0.5919	
		9	0.0224	0.0011	0.0026		0.5833	0.0650	0.0029	0.0076			
733	6	7	0.900	-3.05	-33.80	0.1658	-0.00	-0.10	0.08	15.09	0.1446	0.6620	
		8	0.0143	0.0004	0.0017	0.3160	0.5921	-0.0207	-0.006	-0.0027	0.2201	0.6011	
		9	0.0167	0.0010	0.0020		0.6192	0.0631	0.0027	0.0075			
733	7	7	0.898	-3.05	-22.50	0.1600	-0.01	-0.10	0.07	15.08	0.1756	0.6882	
		8	0.0091	0.0002	0.0011	0.3553	0.6241	-0.0233	-0.008	-0.0031	0.2143	0.6097	
		9	0.0113	0.0008	0.0013		0.6107	0.0619	0.0026	0.0075			

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																			
733	8	7	0.839	-3.04	-11.30	0.1228	-0.03	-0.10	0.07	15.08	0.1926	0.6619	0.0036	0.0000	0.0005	0.7310	-0.0250	-0.0009	-0.0033	0.2105	0.6157	
		8	0.0064	0.0005	0.0008	0.4494	0.6392	0.0607	0.0025	0.0074												
		9	0.0064	0.0005	0.0008	0.4494	0.6392	0.0607	0.0025	0.0074												
733	9	7	0.898	-3.04	-0.00	0.6313	-0.04	-0.10	0.05	15.08	0.2109	0.6691	-0.0011	0.0000	0.0001	-0.0138	-0.0253	-0.0010	-0.0033	0.2071	0.6198	
		8	0.0020	0.0002	0.0001	0.7448	0.3742	0.0605	0.0025	0.0075												
		9	0.0020	0.0002	0.0001	0.7448	0.3742	0.0605	0.0025	0.0075												
733	10	7	0.898	-3.04	11.29	0.2755	-0.04	-0.11	0.04	15.08	0.2161	0.6641	-0.0062	-0.0003	-0.0002	-0.0964	-0.0247	-0.0010	-0.0032	0.1995	0.6175	
		8	0.0021	0.0000	-0.0002	-0.1868	0.5045	0.0614	0.0024	0.0075												
		9	0.0021	0.0000	-0.0002	-0.1868	0.5045	0.0614	0.0024	0.0075												
733	11	7	0.899	-3.05	22.49	0.2976	-0.05	-0.10	0.04	15.08	0.2186	0.6607	-0.0114	-0.0006	-0.0004	-0.6303	-0.0231	-0.0010	-0.0030	0.1950	0.6117	
		8	0.0054	-0.0002	-0.0008	0.2247	0.7329	0.0630	0.0024	0.0077												
		9	0.0054	-0.0002	-0.0008	0.2247	0.7329	0.0630	0.0024	0.0077												
733	12	7	0.899	-3.05	33.79	0.2522	-0.05	-0.10	0.03	15.08	0.2216	0.6676	-0.0175	-0.0008	-0.0020	-0.5932	-0.0206	-0.0009	-0.0027	0.1886	0.6060	
		8	0.0099	-0.0004	-0.0015	0.2415	0.7522	0.0653	0.0024	0.0079												
		9	0.0099	-0.0004	-0.0015	0.2415	0.7522	0.0653	0.0024	0.0079												
733	13	7	0.898	-3.06	45.00	0.2156	-0.07	-0.09	0.02	15.08	0.2148	0.6801	-0.0230	-0.0009	-0.0026	-0.5785	-0.0171	-0.0007	-0.0023	0.1850	0.6015	
		8	0.0155	-0.0006	-0.0020	0.2051	0.6448	0.0682	0.0025	0.0082												
		9	0.0155	-0.0006	-0.0020	0.2051	0.6448	0.0682	0.0025	0.0082												
733	14	7	0.900	-3.07	56.30	0.2497	-0.07	-0.08	0.01	15.08	0.2155	0.7288	-0.0259	-0.0012	-0.0023	-0.6274	-0.0131	-0.0005	-0.0019	0.1828	0.5754	
		8	0.0202	-0.0007	-0.0023	0.1951	0.5760	0.0714	0.0026	0.0085												
		9	0.0202	-0.0007	-0.0023	0.1951	0.5760	0.0714	0.0026	0.0085												
733	15	7	0.898	-3.08	67.50	0.2018	-0.07	-0.08	0.00	15.09	0.2110	0.7762	-0.0298	-0.0012	-0.0035	-0.5939	-0.0084	-0.0003	-0.0013	0.1765	0.5865	
		8	0.0217	-0.0010	-0.0027	0.2426	0.6409	0.0756	0.0026	0.0088												
		9	0.0217	-0.0010	-0.0027	0.2426	0.6409	0.0756	0.0026	0.0088												
733	16	7	0.896	-3.09	78.80	0.2149	-0.07	-0.07	0.01	15.09	0.3244	1.0716	-0.0313	-0.0012	-0.0030	-0.6166	-0.0029	-0.0001	-0.0006	0.1747	0.5607	
		8	0.0227	-0.0012	-0.0030	0.2703	0.6696	0.0793	0.0027	0.0092												
		9	0.0227	-0.0012	-0.0030	0.2703	0.6696	0.0793	0.0027	0.0092												
733	17	7	0.898	-3.10	90.00	0.2000	-0.08	-0.06	0.00	15.10	0.2885	0.5651	-0.0326	-0.0013	-0.0040	-0.6309	-0.0012	0.0000	0.0000	0.1719	0.5671	
		8	0.0241	-0.0012	-0.0030	0.2571	0.6329	0.0825	0.0028	0.0093												
		9	0.0241	-0.0012	-0.0030	0.2571	0.6329	0.0825	0.0028	0.0093												
734	1	7	0.899	-0.01	89.99	0.1663	-0.03	-0.06	0.06	15.10	0.2842	2.6284	-0.0338	-0.0001	-0.0002	-0.3092	-0.0005	-0.0000	-0.0002	0.1931	0.5709	
		8	0.0026	0.0003	0.0002	0.7155	0.5640	0.0789	0.0030	0.0090												
		9	0.0026	0.0003	0.0002	0.7155	0.5640	0.0789	0.0030	0.0090												

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key											
734	2	7	0.897	0.02	78.79	0.1937	-0.03	-0.06	0.06	15.09	-0.0228	2.0734	
		8	-0.0035	-0.0001	-0.0002	1.1120	0.3486	-0.0005	0.0000	-0.0002	0.1934	0.5720	
		9	0.0021	0.0004	0.0004		1.0207	0.0790	0.0030	0.0090			
734	3	7	0.898	0.02	67.49	0.3050	-0.03	-0.06	0.06	15.10	24.6039	0.5671	
		8	-0.0031	-0.0001	-0.0002	0.7348	0.4395	0.0000	0.0000	-0.0001	0.1909		
		9	0.0028	0.0004	0.0003		0.6031	0.0794	0.0030	0.0090			
734	4	7	0.898	0.03	56.29	0.2804	-0.03	-0.06	0.06	15.09	0.5233	-0.4410	
		8	-0.0028	-0.0001	-0.0002	0.7744	0.3859	0.0007	0.0000	-0.0000	0.1923	0.5692	
		9	0.0027	0.0004	0.0003		0.6453	0.0792	0.0030	0.0090			
734	5	7	0.896	0.03	44.99	0.3767	-0.03	-0.06	0.05	15.10	0.5676	0.3453	
		8	-0.0019	-0.0001	-0.0000	0.9380	0.2441	0.0014	0.0001	0.0000	0.1930	0.5706	
		9	0.0024	0.0004	0.0004		0.8657	0.0793	0.0030	0.0090			
734	6	7	0.897	0.04	33.79	0.4477	-0.03	-0.06	0.05	15.10	0.4212	0.4424	
		8	-0.0011	-0.0001	0.0000	1.0454	0.1843	0.0020	0.0001	0.0001	0.1909	0.5676	
		9	0.0022	0.0004	0.0004		0.9490	0.0794	0.0030	0.0090			
734	7	7	0.898	0.05	22.49	0.4614	-0.03	-0.05	0.05	15.10	0.4957	0.5532	
		8	-0.0005	-0.0000	0.0001	0.8918	1.5319	0.0025	0.0002	0.0002	0.1934	0.5726	
		9	0.0024	0.0004	0.0004		0.8292	0.0789	0.0030	0.0090			
734	8	7	0.898	0.06	11.29	-1.3049	-0.02	-0.05	0.06	15.10	0.4326	0.5208	
		8	0.0002	-0.0000	0.0001	1.1493	3.5632	0.0030	0.0002	0.0003	0.1928	0.5712	
		9	0.0019	0.0004	0.0003		1.0284	0.0790	0.0030	0.0090			
734	9	7	0.897	0.06	-0.00	-0.6755	-0.02	-0.05	0.06	15.10	0.4431	0.5626	
		8	0.0004	-0.0000	0.0001	1.1318	1.9729	0.0032	0.0002	0.0003	0.1896	0.5660	
		9	0.0018	0.0004	0.0003		0.9217	0.0796	0.0030	0.0090			
734	10	7	0.898	0.06	-11.30	-1.8795	-0.02	-0.06	0.05	15.09	0.4593	0.5670	
		8	0.0001	-0.0000	0.0001	0.9662	7.2203	0.0033	0.0003	0.0003	0.1929	0.5706	
		9	0.0017	0.0003	0.0002		0.6704	0.0790	0.0030	0.0090			
734	11	7	0.898	0.06	-22.50	0.0973	-0.02	-0.05	0.05	15.09	0.4397	0.5407	
		8	0.0007	-0.0000	0.0000	0.7400	0.4265	0.0034	0.0003	0.0003	0.1932	0.5695	
		9	0.0020	0.0003	0.0001		0.4237	0.0791	0.0030	0.0090			
734	12	7	0.896	0.04	-33.80	0.2525	-0.03	-0.06	0.05	15.10	0.4387	0.5305	
		8	-0.0016	-0.0000	-0.0001	0.7720	0.4623	0.0035	0.0003	0.0003	0.1922	0.5698	
		9	0.0017	0.0002	0.0001		0.4020	0.0798	0.0030	0.0090			

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																							
734	13	7	0.898	-0.0022	0.05	-45.00	0.3355	-0.03	-0.06	0.05	15.10	0.4038	0.5130	7	0.897	-0.0024	0.04	-56.30	0.4454	-0.03	-0.06	0.05	15.10	0.3851	0.5033
		8	-0.0010	0.0003	-0.0002	0.0003	1.4464	0.9723	0.0800	0.0030	0.0090	0.1906	0.5673	9	-0.0012	0.0003	-0.0001	0.0004	1.2681	0.7801	0.0808	0.0039	0.0091	0.1881	0.5628
734	14	7	0.898	-0.0035	0.03	-67.50	0.2987	-0.03	-0.05	0.05	15.09	0.3698	0.4992	7	0.897	-0.0024	0.04	-56.30	0.4454	-0.03	-0.06	0.05	15.10	0.3851	0.5033
		8	-0.0017	0.0002	-0.0001	0.0001	0.8116	0.3688	0.0815	0.0030	0.0091	0.1868	0.5623	9	-0.0017	0.0002	-0.0001	0.0001	0.8116	0.3688	0.0815	0.0039	0.0091	0.1868	0.5623
734	15	7	0.897	-0.0033	0.03	-78.80	0.4522	-0.03	-0.06	0.05	15.10	0.3274	0.4387	7	0.897	-0.0033	0.03	-78.80	0.4522	-0.03	-0.06	0.05	15.10	0.3274	0.4387
		8	-0.0001	0.0004	-0.0003	-18.0999	-15.0438	0.9477	0.0817	0.0030	0.0091	0.1864	0.5610	9	-0.0001	0.0004	-0.0003	-18.0999	-15.0438	0.9477	0.0817	0.0030	0.0091	0.1864	0.5610
734	16	7	0.896	-0.0017	0.03	-90.00	0.4032	-0.03	-0.06	0.05	15.10	0.3275	0.4077	7	0.896	-0.0017	0.03	-90.00	0.4032	-0.03	-0.06	0.05	15.10	0.3275	0.4077
		8	-0.0037	0.0002	-0.0006	0.0006	0.8152	0.4506	0.0822	0.0030	0.0092	0.1856	0.5616	9	-0.0037	0.0002	-0.0006	0.0006	0.8152	0.4506	0.0822	0.0030	0.0092	0.1856	0.5616
735	1	7	0.896	-0.0251	3.16	-89.99	0.1872	-0.08	-0.05	0.00	15.10	0.3250	0.4245	7	0.896	-0.0251	3.16	-89.99	0.1872	-0.08	-0.05	0.00	15.10	0.3250	0.4245
		8	-0.0361	-0.0012	-0.0045	0.0045	0.2569	0.6200	0.0850	0.0028	0.0096	0.1695	0.5646	9	-0.0361	-0.0012	-0.0045	0.0045	0.2569	0.6200	0.0850	0.0028	0.0096	0.1695	0.5646
735	2	7	0.897	-0.0337	3.16	-78.79	0.1958	-0.08	-0.05	0.01	15.10	0.3150	0.6104	7	0.897	-0.0337	3.16	-78.79	0.1958	-0.08	-0.05	0.01	15.10	0.3150	0.6104
		8	-0.0227	-0.0012	-0.0030	0.0030	0.2766	0.6639	0.0881	0.0029	0.0100	0.1671	0.5628	9	-0.0227	-0.0012	-0.0030	0.0030	0.2766	0.6639	0.0881	0.0029	0.0100	0.1671	0.5628
735	3	7	0.898	-0.0299	3.17	-67.49	0.1973	-0.07	-0.04	0.00	15.10	0.2527	0.6483	7	0.898	-0.0299	3.17	-67.49	0.1973	-0.07	-0.04	0.00	15.10	0.2527	0.6483
		8	-0.0209	-0.0011	-0.0027	0.0027	0.2645	0.6458	0.0902	0.0029	0.0100	0.1645	0.5578	9	-0.0209	-0.0011	-0.0027	0.0027	0.2645	0.6458	0.0902	0.0029	0.0100	0.1645	0.5578
735	4	7	0.896	-0.0175	3.17	-56.29	0.1790	-0.06	-0.03	0.01	15.10	0.2494	0.6380	7	0.896	-0.0175	3.17	-56.29	0.1790	-0.06	-0.03	0.01	15.10	0.2494	0.6380
		8	-0.0258	-0.0009	-0.0024	0.0024	0.2763	0.6899	0.0931	0.0029	0.0104	0.1583	0.5579	9	-0.0258	-0.0009	-0.0024	0.0024	0.2763	0.6899	0.0931	0.0029	0.0104	0.1583	0.5579
735	5	7	0.895	-0.0208	3.17	-44.99	0.1611	-0.05	-0.02	0.03	15.10	0.2477	0.6383	7	0.895	-0.0208	3.17	-44.99	0.1611	-0.05	-0.02	0.03	15.10	0.2477	0.6383
		8	-0.0135	-0.0007	-0.0020	0.0020	0.2857	0.7589	0.0942	0.0029	0.0104	0.1577	0.5548	9	-0.0135	-0.0007	-0.0020	0.0020	0.2857	0.7589	0.0942	0.0029	0.0104	0.1577	0.5548
735	6	7	0.896	-0.0151	3.17	-33.79	0.1458	-0.05	-0.01	0.03	15.11	0.2422	0.6386	7	0.896	-0.0151	3.17	-33.79	0.1458	-0.05	-0.01	0.03	15.11	0.2422	0.6386
		8	-0.0081	-0.0005	-0.0015	0.0015	0.3510	0.9799	0.0953	0.0029	0.0105	0.1558	0.5529	9	-0.0081	-0.0005	-0.0015	0.0015	0.3510	0.9799	0.0953	0.0029	0.0105	0.1558	0.5529

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																			
735	7	0.897	3.17	-22.50	0.2126	-0.04	-0.01	0.04	15.11	0.2360	0.6442	0.0082	-0.0003	-0.0011	0.1630	0.6890	0.323	0.0015	0.0041	0.1565	0.5508
	8	-0.0054	-0.0001	-0.0007	0.1630	0.7296	0.0951	0.0029	0.0041												
	9																				
735	8	0.897	3.17	-11.30	0.0557	-0.03	-0.00	0.05	15.10	0.2247	0.6431	-0.0035	-0.0000	-0.0004	-0.4452	0.6383	0.0015	0.0043	0.1580	0.5500	
	8	-0.0011	0.0001	-0.0002	-0.4452	1.0689	0.0947	0.0029	0.0043												
	9																				
735	9	0.898	3.17	-0.00	0.9579	-0.02	-0.01	0.07	15.11	0.2153	0.6455	0.0009	0.0001	0.0002	0.9579	1.4595	0.0014	0.0043	0.1609	0.5523	
	8	0.0020	0.0005	0.0005	1.2701	1.3281	0.0938	0.0030	0.0103												
	9																				
735	10	0.897	3.17	11.29	0.2612	-0.01	-0.00	0.07	15.11	0.2050	0.6471	0.0068	0.0003	0.0010	0.4346	0.8036	0.0013	0.0042	0.1638	0.5522	
	8	0.0068	0.0007	0.0010	0.4346	0.6122	0.0930	0.0030	0.0102												
	9																				
735	11	0.896	3.16	22.49	0.0955	-0.01	-0.02	0.09	15.10	0.1887	0.6461	0.0142	0.0002	0.0016	0.0955	0.5910	0.0011	0.0038	0.1654	0.5500	
	8	0.0156	0.0010	0.0017	0.3386	0.5554	0.0921	0.0030	0.0101												
	9																				
735	12	0.898	3.16	33.79	0.1541	-0.00	-0.03	0.10	15.11	0.1786	0.6494	0.0186	0.0005	0.0023	0.1541	0.6392	0.0009	0.0033	0.1713	0.5550	
	8	0.0196	0.0014	0.0025	0.3577	0.6392	0.0901	0.0030	0.0100												
	9																				
735	13	0.898	3.15	44.99	0.1724	0.00	-0.04	0.10	15.10	0.1680	0.6483	0.0223	0.0007	0.0028	0.1724	0.6437	0.0007	0.0027	0.1751	0.5520	
	8	0.0252	0.0015	0.0031	0.3160	0.6289	0.0880	0.0030	0.0097												
	9																				
735	14	0.898	3.15	56.29	0.2004	0.00	-0.04	0.11	15.10	0.1671	0.6534	0.0254	0.0009	0.0033	0.2004	0.6513	0.0005	0.0020	0.1635	0.5554	
	8	0.0291	0.0017	0.0037	0.3069	0.6434	0.0857	0.0031	0.0095												
	9																				
735	15	0.898	3.14	67.49	0.2711	0.01	-0.05	0.11	15.09	0.3539	0.5958	0.0269	0.0010	0.0039	0.2711	0.6560	0.0005	0.0006	0.1696	0.5536	
	8	0.0332	0.0018	0.0039	0.2711	0.5901	0.0823	0.0031	0.0091												
	9																				
735	16	0.895	3.14	78.79	0.2116	0.01	-0.05	0.12	15.10	0.4205	0.4399	0.0271	0.0011	0.0042	0.2116	0.6593	0.0002	0.0002	0.1981	0.5551	
	8	0.0336	0.0019	0.0042	0.2847	0.6553	0.0787	0.0031	0.0087												
	9																				
735	17	0.897	3.14	89.99	0.2353	0.01	-0.07	0.12	15.10	1.5008	3.9634	0.0259	0.0012	0.0043	0.2353	0.6827	0.0000	0.0002	1.5008	3.9634	
	8	0.0321	0.0019	0.0043	0.3015	0.6827	0.0763	0.0031	0.0085												
	9																				

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key															
736	1	7	0.897	6.26	89.99	0.2168	0.6319	0.06	-0.06	0.010	0.16	15.10	-0.1398	-0.0472				
		8	0.0584	0.0025	0.0073	0.2606	0.6019	0.0727	-0.0010	0.0000	0.0032	0.0001	0.2202	0.5600				
		9	0.0588	0.0030	0.0070													
736	2	7	0.896	6.26	78.79	0.2033	0.6390	0.06	-0.05	0.0099	0.16	15.10	0.1984	0.6329				
		8	0.0594	0.0024	0.0076	0.2663	0.6016	0.0790	0.0005	0.0031	0.0087	0.0000	0.1991	0.5506				
		9	0.0583	0.0031	0.0070													
736	3	7	0.898	6.26	67.49	0.1921	0.6525	0.06	-0.03	0.0230	0.16	15.10	0.1232	0.6375				
		8	0.0574	0.0022	0.0074	0.2738	0.6005	0.0655	0.0005	0.0031	0.0029	0.0094	0.1652	0.5496				
		9	0.0560	0.0030	0.0067													
736	4	7	0.898	6.26	56.29	0.1770	0.6436	0.05	-0.01	0.0345	0.15	15.10	0.1332	0.6343				
		8	0.0528	0.0018	0.0068	0.2923	0.6220	0.0693	0.0009	0.0029	0.0009	0.0043	0.1650	0.5505				
		9	0.0506	0.0029	0.0063													
736	5	7	0.899	6.26	44.99	0.1748	0.6520	0.03	-0.00	0.0451	0.13	15.10	0.1485	0.6461				
		8	0.0452	0.0015	0.0059	0.2814	0.5951	0.0925	0.0000	0.0027	0.0013	0.0058	0.1482	0.5501				
		9	0.0457	0.0025	0.0054													
736	6	7	0.896	6.26	33.79	0.1566	0.6229	0.02	0.00	0.0524	0.12	15.10	0.1647	0.6367				
		8	0.0363	0.0011	0.0045	0.3113	0.6381	0.0962	0.0000	0.0026	0.0017	0.0066	0.1370	0.5501				
		9	0.0363	0.0022	0.0046													
736	7	7	0.900	6.27	22.49	0.1524	0.6063	0.00	0.02	0.0570	0.11	15.10	0.1821	0.6234				
		8	0.0250	0.0008	0.0030	0.3418	0.6638	0.0978	0.0000	0.0020	0.0020	0.0071	0.1311	0.5544				
		9	0.0263	0.0017	0.0034													
736	8	7	0.897	6.27	11.29	0.1786	0.6184	0.00	0.02	0.0599	0.08	15.10	0.1975	0.6110				
		8	0.0134	0.0004	0.0016	0.3565	0.5864	0.1004	0.0000	0.0023	0.0025	0.0073	0.1260	0.5558				
		9	0.0159	0.0011	0.0018													
736	9	7	0.896	6.27	-0.00	0.1605	0.7159	0.02	0.03	0.0605	0.06	15.10	0.2116	0.6084				
		8	0.0027	0.0000	0.0004	0.6463	0.7145	0.1019	0.0000	0.0025	0.0025	0.0073	0.1239	0.5572				
		9	0.0040	0.0005	0.0005													
736	10	7	0.897	6.27	-11.30	0.3260	0.6361	0.03	0.03	0.0596	0.04	15.10	0.2196	0.6048				
		8	-0.0064	-0.0004	-0.0008	0.1624	0.6698	0.1028	0.0000	0.0026	0.0024	0.0072	0.1212	0.5531				
		9	-0.0047	-0.0001	-0.0006													
736	11	7	0.897	6.28	-22.50	0.2155	0.5905	0.02	0.02	0.0570	0.02	15.10	0.2279	0.6060				
		8	-0.0179	-0.0007	-0.0021	0.2167	0.5616	0.1030	0.0000	0.0026	0.0025	0.0069	0.1232	0.5548				
		9	-0.0157	-0.0006	-0.0017													

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key												
736	12	7	0.897	6.27	-33.79	0.2012	-0.07	0.01	-0.00	15.10	0.2381	0.6114			
	8	8	-0.0297	-0.0011	-0.0037	0.2900	0.6219	0.0521	0.0024	0.0063	0.1285	0.5559			
	9	9	-0.0242	-0.0014	-0.0031		0.6542	0.1028	0.0026	0.0114					
736	13	7	0.897	6.27	-44.99	0.2007	-0.08	0.01	-0.01	15.11	0.2441	0.6232			
	8	8	-0.0408	-0.0016	-0.0053	0.2816	0.6570	0.0458	0.0022	0.0057	0.1374	0.5560			
	9	9	-0.0340	-0.0019	-0.0042		0.6283	0.1033	0.0028	0.0114					
736	14	7	0.898	6.27	-56.29	0.1874	-0.10	-0.00	-0.02	15.11	0.2540	0.6292			
	8	8	-0.0507	-0.0019	-0.0065	0.2771	0.6429	0.0364	0.0018	0.0045	0.1430	0.5572			
	9	9	-0.0421	-0.0023	-0.0052		0.6273	0.1010	0.0028	0.0112					
736	15	7	0.897	6.27	-67.49	0.1846	-0.11	-0.01	-0.04	15.11	0.2560	0.6341			
	8	8	-0.0579	-0.0021	-0.0072	0.2671	0.6277	0.0259	0.0013	0.0032	0.1449	0.5627			
	9	9	-0.0486	-0.0025	-0.0059		0.6102	0.0987	0.0028	0.0111					
736	16	7	0.897	6.27	-78.79	0.1984	-0.12	-0.04	-0.04	15.10	0.2517	0.6230			
	8	8	-0.0609	-0.0024	-0.0075	0.2532	0.6163	0.0154	0.0007	0.0018	0.1473	0.5677			
	9	9	-0.0534	-0.0027	-0.0062		0.5897	0.0935	0.0027	0.0106					
736	17	7	0.897	6.27	-89.99	0.2262	-0.13	-0.05	-0.05	15.09	0.2702	0.4629			
	8	8	-0.0611	-0.0027	-0.0076	0.2427	0.6230	0.0042	0.0002	0.0003	0.1486	0.5727			
	9	9	-0.0562	-0.0027	-0.0064		0.5754	0.0867	0.0025	0.0099					
737	1	7	0.896	9.41	-90.00	0.1898	-0.16	-0.06	-0.07	15.09	0.2253	0.4614			
	8	8	-0.0861	-0.0032	-0.0099	0.2158	0.5795	0.0037	0.0001	0.0003	0.1101	0.5924			
	9	9	-0.0790	-0.0034	-0.0088		0.5596	0.0926	0.0020	0.0109					
737	2	7	0.897	9.42	-78.79	0.1853	-0.16	-0.03	-0.06	15.10	0.2564	0.6255			
	8	8	-0.0862	-0.0031	-0.0102	0.2268	0.5959	0.0193	0.0009	0.0024	0.1253	0.5705			
	9	9	-0.0764	-0.0034	-0.0084		0.5550	0.1013	0.0025	0.0115					
737	3	7	0.897	9.42	-67.48	0.1672	-0.15	-0.00	-0.06	15.11	0.2645	0.6280			
	8	8	-0.0831	-0.0027	-0.0099	0.2573	0.5973	0.0357	0.0018	0.0044	0.1262	0.5640			
	9	9	-0.0691	-0.0035	-0.0079		0.5771	0.1081	0.0027	0.0121					
737	4	7	0.898	9.41	-56.29	0.1687	-0.13	0.01	-0.06	15.11	0.2541	0.6051			
	8	8	-0.0746	-0.0025	-0.0092	0.2710	0.6200	0.0500	0.0025	0.0060	0.1243	0.5576			
	9	9	-0.0621	-0.0033	-0.0069		0.5623	0.1122	0.0027	0.0125					
737	5	7	0.999	9.41	-44.99	0.1697	-0.12	0.03	-0.05	15.11	0.2422	0.5924			
	8	8	-0.0621	-0.0021	-0.0077	0.2863	0.6275	0.0610	0.0029	0.0072	0.1140	0.5569			
	9	9	-0.0528	-0.0030	-0.0060		0.5740	0.1110	0.0025	0.0123					

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd																					
737	6	7	0.897	9.40	-33.79	0.1937	-0.09	0.04	-0.01	15.10	0.2276	0.5761											
	8	8	-0.0451	-0.0017	-0.0057	0.2742	0.6417	0.0692	0.0031	0.0123	0.1080	0.5524											
	9	9	-0.0419	-0.0023	-0.0048		0.5823	0.1114	0.0024														
737	7	7	0.898	9.41	-22.50	0.2142	-0.07	0.05	0.00	15.11	0.2090	0.5751											
	8	8	-0.0273	-0.0011	-0.0033	0.2803	0.6053	0.0762	0.0031	0.0122	0.1147	0.5488											
	9	9	-0.0263	-0.0014	-0.0031		0.6005	0.1116	0.0025														
737	8	7	0.898	9.42	-11.30	0.2853	-0.04	0.06	0.03	15.11	0.1943	0.5748											
	8	8	-0.0101	-0.0005	-0.0011	0.2706	0.5619	0.0806	0.0031	0.0122	0.1149	0.5450											
	9	9	-0.0099	-0.0005	-0.0012		0.6449	0.1121	0.0025														
737	9	7	0.898	9.41	0.00	0.0454	-0.02	0.06	0.06	15.11	0.1870	0.5792											
	8	8	0.0043	0.0000	0.0005	1.0373	0.5377	0.0829	0.0031	0.0122	0.1148	0.5500											
	9	9	0.0025	0.0005	0.0007		1.4075	0.1116	0.0025														
737	10	7	0.898	9.41	11.30	0.1680	0.00	0.04	0.10	15.11	0.1805	0.5842											
	8	8	0.0203	0.0006	0.0025	0.3869	0.6308	0.0829	0.0029	0.0120	0.1668	0.5492											
	9	9	0.0189	0.0014	0.0027		0.7110	0.1099	0.0025														
737	11	7	0.897	9.40	22.50	0.1685	0.02	0.05	0.13	15.10	0.1695	0.5920											
	8	8	0.0369	0.0012	0.0048	0.3083	0.6564	0.0805	0.0027	0.0116	0.1221	0.5479											
	9	9	0.0367	0.0022	0.0045		0.6175	0.1066	0.0026														
737	12	7	0.898	9.39	33.79	0.1535	0.04	0.03	0.16	15.10	0.1544	0.6019											
	8	8	0.0543	0.0016	0.0071	0.2983	0.6340	0.0754	0.0023	0.0112	0.1287	0.5494											
	9	9	0.0502	0.0029	0.0060		0.6046	0.1025	0.0026														
737	13	7	0.897	9.39	44.99	0.1493	0.07	0.01	0.18	15.10	0.1413	0.6175											
	8	8	0.0687	0.0020	0.0087	0.2868	0.6391	0.0656	0.0018	0.0081	0.1292	0.5321											
	9	9	0.0607	0.0034	0.0072		0.5939	0.0983	0.0025														
737	14	7	0.898	9.39	56.29	0.1736	0.08	-0.00	0.18	15.10	0.1426	0.6283											
	8	8	0.0743	0.0025	0.0091	0.2476	0.6121	0.0507	0.0014	0.0102	0.1404	0.5502											
	9	9	0.0713	0.0035	0.0080		0.5611	0.0934	0.0026														
737	15	7	0.898	9.39	67.48	0.1810	0.08	-0.02	0.19	15.10	0.1504	0.6067											
	8	8	0.0780	0.0028	0.0093	0.2359	0.5960	0.0331	0.0009	0.0047	0.1567	0.5510											
	9	9	0.0764	0.0036	0.0087		0.5726	0.0881	0.0029														
737	16	7	0.897	9.39	78.79	0.1897	0.09	-0.05	0.20	15.10	0.1538	0.5794											
	8	8	0.0789	0.0029	0.0091	0.2254	0.5815	0.0154	0.0004	0.0090	0.1595	0.5521											
	9	9	0.0801	0.0036	0.0092		0.5771	0.0821	0.0032														

See Key →

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd																		
737	17	7	0.897	9.39	89.99	0.1992	0.09	-0.06	0.19	15.09	-0.0648	-1.7711								
		8	0.0783	0.0031	0.0090	0.2158	0.5801	0.0709	-0.0000	-0.0001	0.2287	0.5608								
		9	0.0819	0.0035	0.0094	0.1992	0.5736	0.0709	-0.0032	0.0079	0.2287	0.5608								
738	1	7	0.897	12.50	89.99	0.1582	0.11	-0.07	0.20	15.10	0.2319	-3.3694								
		8	0.0968	0.0030	0.0108	0.1719	0.5610	0.0002	0.0000	-0.0001	0.2293	0.5566								
		9	0.0981	0.0033	0.0112	0.1719	0.5752	0.0701	0.0032	0.0078	0.2293	0.5566								
73A	2	7	0.897	12.50	78.78	0.1640	0.11	-0.04	0.19	15.10	0.1441	0.5695								
		8	0.0983	0.0032	0.0112	0.1676	0.5715	0.209	0.0006	0.0023	0.1441	0.5695								
		9	0.0955	0.0032	0.0105	0.1676	0.5544	0.0825	0.0031	0.0090	0.1441	0.5695								
738	3	7	0.897	12.49	67.48	0.1538	0.11	-0.01	0.19	15.10	0.1396	0.6203								
		8	0.0994	0.0030	0.0113	0.1923	0.5723	0.0441	0.0012	0.0054	0.1396	0.6203								
		9	0.0889	0.0034	0.0102	0.1923	0.5737	0.0913	0.0027	0.0100	0.1396	0.6203								
73A	4	7	0.899	12.49	56.29	0.1476	0.10	0.01	0.19	15.10	0.1265	0.5355								
		8	0.0954	0.0028	0.0113	0.2172	0.5936	0.0671	0.0016	0.0085	0.1265	0.5355								
		9	0.0828	0.0036	0.0091	0.2172	0.5550	0.0971	0.0025	0.0106	0.1265	0.5355								
738	5	7	0.899	12.49	44.99	0.1393	0.09	0.04	0.19	15.10	0.1255	0.6214								
		8	0.0878	0.0024	0.0109	0.2477	0.6263	0.0858	0.0021	0.0106	0.1255	0.6214								
		9	0.0753	0.0037	0.0083	0.2477	0.5544	0.1023	0.0025	0.0113	0.1255	0.6214								
738	6	7	0.899	12.50	33.79	0.1335	0.06	0.05	0.17	15.10	0.1362	0.5998								
		8	0.0716	0.0019	0.0092	0.2924	0.6459	0.0971	0.0026	0.0120	0.1362	0.5998								
		9	0.0605	0.0035	0.0069	0.2924	0.5737	0.1077	0.0023	0.0120	0.1362	0.5998								
73A	7	7	0.899	12.50	22.51	0.1480	0.03	0.06	0.14	15.10	0.1458	0.5853								
		8	0.0490	0.0014	0.0062	0.3087	0.6409	0.1015	0.0029	0.0118	0.1458	0.5853								
		9	0.0447	0.0027	0.0053	0.3087	0.6010	0.1122	0.0022	0.0125	0.1458	0.5853								
73A	8	7	0.899	12.51	11.31	0.1450	0.01	0.07	0.11	15.10	0.1523	0.5757								
		8	0.0262	0.0007	0.0031	0.3438	0.6026	0.1034	0.0031	0.0119	0.1523	0.5757								
		9	0.0239	0.0016	0.0031	0.3438	0.6463	0.1156	0.0022	0.0129	0.1523	0.5757								
738	9	7	0.899	12.52	0.00	0.0882	-0.02	0.07	0.06	15.11	0.1528	0.5652								
		8	0.0053	0.0000	0.0006	0.6982	0.5922	0.1012	0.0030	0.0114	0.1528	0.5652								
		9	0.0029	0.0004	0.0004	0.6982	0.8078	0.1176	0.0021	0.0131	0.1528	0.5652								
73A	10	7	0.897	12.51	-11.30	0.2489	-0.05	0.06	0.02	15.11	0.1592	0.5628								
		8	-0.0138	-0.0006	-0.0015	0.3009	0.5653	0.0967	0.0030	0.0108	0.1592	0.5628								
		9	-0.0152	-0.0009	-0.0020	0.3009	0.6682	0.1203	0.0021	0.0134	0.1592	0.5628								

See Key

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																												
738	11	7	0.898	12.51	-22.51	0.2007	-0.08	0.06	-0.02	15.11	0.1719	0.5573	8	-0.369	-0.0014	-0.0046	0.6344	0.0904	0.0031	0.0100	0.0928	0.5529	9	-0.0374	-0.0020	-0.0045	0.6012	0.1214	0.0022	0.0134
73A	12	7	0.901	12.52	-33.79	0.1530	-0.11	0.06	-0.05	15.11	0.1972	0.5556	8	-0.618	-0.0018	-0.0078	0.6323	0.0824	0.0032	0.0091	0.1063	0.5447	9	-0.0526	-0.0031	-0.0060	0.5706	0.1194	0.0025	0.0130
73A	13	7	0.897	12.52	-44.99	0.1409	-0.14	0.05	-0.07	15.11	0.2235	0.5614	8	-0.827	-0.0023	-0.0103	0.6249	0.0745	0.0033	0.0083	0.0977	0.5535	9	-0.0657	-0.0034	-0.0075	0.5737	0.1204	0.0023	0.0133
73A	14	7	0.898	12.52	-56.28	0.1506	-0.16	0.04	-0.08	15.12	0.2588	0.5718	8	-0.969	-0.0029	-0.0118	0.6121	0.0607	0.0031	0.0069	0.1077	0.5625	9	-0.0786	-0.0033	-0.0084	0.5392	0.1240	0.0026	0.0139
73A	15	7	0.901	12.53	-67.48	0.1784	-0.17	0.00	-0.09	15.11	0.2698	0.6063	8	-1.046	-0.0032	-0.0124	0.5948	0.0437	0.0023	0.0053	0.1096	0.5694	9	-0.0866	-0.0030	-0.0091	0.5297	0.1176	0.0025	0.0133
73A	16	7	0.898	12.54	-78.80	0.1634	-0.18	0.03	-0.09	15.10	0.2662	0.6218	8	-1.071	-0.0035	-0.0124	0.5817	0.0233	0.0012	0.0029	0.1004	0.6218	9	-0.0904	-0.0030	-0.0099	0.5496	0.1107	0.0022	0.0128
73A	17	7	0.898	12.53	-90.01	0.1621	-0.18	0.06	-0.08	15.08	0.1534	0.6165	8	-1.061	-0.0034	-0.0119	0.5643	0.0034	0.0001	0.0001	0.0774	0.2880	9	-0.0930	-0.0030	-0.0103	0.5560	0.0993	0.0015	0.0122
739	3	7	0.799	-3.12	-0.00	-0.2534	-0.02	-3.03	0.06	-3.04	0.2653	0.6214	8	0.021	0.0001	0.0002	0.6930	-0.0419	-0.0022	-0.0053	0.3053	0.6352	9	-0.0023	0.0002	-0.0000	0.1975	-0.0397	-0.0024	-0.0049
739	4	7	0.800	-1.08	-0.00	0.0145	-0.02	-2.98	0.06	-3.01	0.2695	0.6615	8	0.023	0.0000	0.0004	0.9459	-0.0217	-0.0011	-0.0028	0.3396	0.6615	9	-0.0009	0.0002	-0.0000	0.5192	-0.0209	-0.0014	-0.0027
739	5	7	0.800	-0.03	-0.00	-0.0503	-0.02	-2.96	0.06	-3.00	0.2518	0.6874	8	0.031	-0.0000	0.0004	0.7275	-0.0125	-0.0006	-0.0017	0.3826	0.6874	9	-0.0004	0.0001	-0.0001	1.7911	-0.0126	-0.0009	-0.0017
739	6	7	0.800	0.48	-0.00	0.0426	-0.02	-2.95	0.06	-2.99	0.2431	0.7090	8	0.031	0.0000	0.0004	0.7901	-0.0079	-0.0003	-0.0011	0.4252	0.7320	9	-0.0007	0.0001	-0.0001	0.9355	-0.0086	-0.0007	-0.0012

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																		
739	7	0.799	0.97	-0.00	-0.02	-2.94	0.06	-2.97	0.2502	0.9576	0.0036	0.0000	0.0004	0.0191	0.6673	-0.0032	-0.0001	0.0001	0.5076	0.7311
	8	-0.0007	0.0000	0.0001	0.9168	-0.0051	-0.0005	-0.0006												
	9																			
739	8	0.800	3.10	-0.00	-0.02	-2.90	0.06	-2.94	0.2997	0.5839	0.0029	0.0000	0.0004	0.0254	0.7570	0.0128	0.0007	0.0003	0.1712	0.6152
	9	-0.0017	0.0002	0.0000	-0.0081	0.0104	0.0003	0.0012												
739	9	0.798	6.20	-0.00	-0.02	-2.85	0.06	-2.88	0.2716	0.6089	0.0015	-0.0001	0.0003	0.0225	1.1894	0.0400	0.0021	0.0019	0.2425	0.6114
	8	0.0009	0.0001	0.0001	-0.5587	0.0398	0.0019	0.0048												
	9																			
739	10	0.797	9.30	-0.00	-0.02	-2.79	0.06	-2.84	0.2743	0.5596	0.0019	0.0002	0.0003	0.3191	0.7801	0.0613	0.0033	0.0031	0.2522	0.5566
	8	-0.0009	0.0002	0.0000	0.0740	0.0630	0.0031	0.0070												
	9																			
739	11	0.797	12.51	-0.00	-0.03	-2.77	0.06	-2.82	0.2153	0.5415	0.0019	0.0002	0.0002	0.4270	0.7154	0.0813	0.0035	0.0033	0.2065	0.5339
	8	-0.0013	0.0001	0.0001	0.6541	0.0814	0.0033	0.0088												
	9																			
739	12	0.800	-0.07	-0.00	-0.01	-2.96	0.06	-2.98	0.2691	0.7137	0.0024	0.0001	0.0005	0.2121	1.1426	0.0123	0.0006	0.0009	0.3748	0.6899
	8	-0.0006	0.0002	0.0001	1.0409	-0.0129	-0.0009	-0.0017												
	9																			
740	1	0.798	-3.13	-0.00	-0.02	-1.01	0.06	-1.05	0.2732	0.6484	0.0020	0.0000	0.0003	0.1521	0.8175	0.0300	0.0016	0.0018	0.3138	0.6339
	8	-0.0021	0.0002	0.0001	0.3146	-0.0290	-0.0018	-0.0036												
	9																			
740	2	0.799	-1.06	-0.00	-0.02	-0.97	0.06	-1.01	0.2607	0.7049	0.0027	0.0000	0.0004	0.0732	0.7823	0.0101	0.0005	0.0007	0.3909	0.6719
	8	-0.0010	0.0002	0.0001	0.5736	-0.0100	-0.0007	-0.0013												
	9																			
740	3	0.799	-0.02	-0.00	-0.01	-0.95	0.06	-0.99	0.2710	1.1169	0.0031	0.0000	0.0004	0.0179	0.7753	0.0018	0.0000	0.0004	0.5550	0.6923
	8	-0.0006	0.0001	0.0001	1.3097	-0.0034	-0.0003	-0.0004												
	9																			
740	4	0.797	0.49	-0.00	-0.02	-0.94	0.06	-0.98	0.2710	0.3453	0.0038	0.0000	0.0004	0.1083	0.5483	0.0021	0.0001	0.0001	0.3673	0.6832
	8	-0.0010	0.0002	0.0001	0.5699	0.0000	-0.0001	-0.0001												
	9																			
740	5	0.797	0.98	-0.00	-0.02	-0.93	0.06	-0.97	0.3302	0.5700	0.0035	0.0000	0.0005	0.0302	0.7035	0.0058	0.0003	0.0000	0.3302	0.5700
	8	-0.0011	0.0002	0.0000	0.3750	0.0040	0.0000	0.0005												
	9																			

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

← See Key →

Run Pt.	Cd																										
740	6	7	8	9	3.09	-0.0004	-0.0246	-0.0001	-0.0004	-0.02	0.6458	0.5165	-0.89	0.239	0.0212	0.06	0.0013	0.0029	-0.0029	0.2787	0.2397	0.6065	0.6268				
					0.799	0.0034	0.0010	0.0002	0.0000	0.246	0.5165	0.0212	0.06	0.0013	0.0029	0.2787	0.2397	0.6065	0.6268								
740	7	7	8	9	6.22	-0.0002	-0.2959	0.0001	-0.0002	-0.02	0.6164	-0.3111	-0.83	0.505	0.0025	0.06	0.0027	0.0060	-0.0060	0.2749	0.2478	0.5950	0.6019				
					0.798	0.0023	0.0006	0.0003	0.0000	0.6164	-0.3111	0.505	0.0025	0.06	0.0027	0.0060	0.0061	0.5950	0.6019								
740	8	7	8	9	9.32	-0.0001	-0.1546	0.0003	-0.0001	-0.02	1.1007	-1.7901	-0.79	0.708	0.0034	0.06	0.0034	0.0078	-0.0078	0.2453	0.2372	0.5543	0.5494				
					0.798	0.0016	0.0003	0.0001	0.0003	1.1007	-1.7901	0.708	0.0034	0.06	0.0034	0.0078	0.0078	0.5543	0.5494								
740	9	7	8	9	12.50	-0.0001	-0.3803	0.0003	-0.0001	-0.02	0.7931	0.3214	-0.79	0.850	0.0031	0.06	0.0030	0.0093	-0.0093	0.1786	0.1812	0.5442	0.5393				
					0.797	0.0019	0.0015	0.0001	0.0003	0.7931	0.3214	0.850	0.0031	0.06	0.0030	0.0093	0.1786	0.1812	0.5442	0.5393							
740	10	7	8	9	-0.04	0.0002	-0.1619	0.0001	0.0002	-0.02	0.4969	0.4080	-0.95	0.014	-0.0003	0.06	0.0003	-0.0004	0.99	0.3228	0.5624	1.2467	0.7124				
					0.797	0.0040	0.0010	0.0002	0.0004	0.4969	0.4080	0.014	-0.0003	0.06	0.0003	-0.0004	0.99	0.3228	0.7124								
741	1	7	8	9	-3.10	0.0000	-0.0277	0.0001	-0.0003	-0.02	1.0623	0.8801	-0.12	0.244	-0.0014	0.06	0.0013	-0.0031	0.07	0.2759	0.3264	0.6492	0.6476				
					0.798	0.0018	0.0011	0.0001	0.0003	1.0623	0.8801	0.244	-0.0014	0.06	0.0013	-0.0031	0.07	0.2759	0.6492								
741	2	7	8	9	-1.07	0.0002	-0.0612	0.0001	-0.0004	-0.02	0.8260	0.8335	-0.07	0.054	-0.0004	0.06	0.0002	-0.0007	0.04	0.2630	0.4700	0.7764	0.6759				
					0.797	0.0025	0.0007	0.0002	0.0004	0.8260	0.8335	0.054	-0.0004	0.06	0.0002	-0.0007	0.04	0.2630	0.6759								
741	3	7	8	9	-0.01	0.0002	-0.1250	0.0001	-0.0004	-0.01	0.5940	0.9869	-0.05	0.025	-0.0001	0.06	0.0001	0.0002	0.02	0.3372	-0.5180	0.4591	0.7678				
					0.797	0.0036	0.0006	0.0002	0.0004	0.5940	0.9869	0.025	-0.0001	0.06	0.0001	0.0002	0.02	0.3372	0.7678								
741	4	7	8	9	0.49	0.0002	0.0927	0.0001	-0.0005	-0.02	0.8653	0.7326	-0.05	0.062	0.0001	0.06	0.0004	0.0006	0.01	0.3296	0.1253	0.5898	0.6934				
					0.797	0.0030	0.0008	0.0002	0.0005	0.8653	0.7326	0.062	0.0001	0.06	0.0004	0.0006	0.01	0.3296	0.6934								
741	5	7	8	9	1.01	0.0000	0.1373	0.0001	-0.0005	-0.02	0.9122	1.0778	-0.103	0.0107	0.0003	0.06	0.0006	0.0013	0.00	0.3055	0.1691	0.6069	0.6395				
					0.798	0.0028	0.0007	0.0001	0.0005	0.9122	1.0778	0.0107	0.0003	0.06	0.0006	0.0013	0.00	0.3055	0.6395								
741	6	7	8	9	3.10	0.0000	0.0312	0.0001	-0.0004	-0.02	0.7271	0.911A	0.000	0.0296	0.0013	0.06	0.0016	0.0033	0.03	0.2794	0.2450	0.6184	0.6216				
					0.797	0.0031	0.0006	0.0002	0.0004	0.7271	0.911A	0.0296	0.0013	0.06	0.0016	0.0033	0.03	0.2794	0.6216								

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																	
741	7	0.799	6.25	-0.00	0.0072	-0.02	0.06	0.06	0.06	0.06	0.06	0.06	0.06	0.06	0.06	0.06	0.06	0.2784	0.5834
	8	0.0015	0.0000	0.0003	0.0072	1.2074	0.0548	0.0557	0.0548	0.0557	0.0548	0.0557	0.0548	0.0557	0.0548	0.0557	0.0548	0.2542	0.5801
	9	0.0011	0.0002	-0.0000	0.9672	-0.3937	0.0557	0.0557	0.0557	0.0557	0.0557	0.0557	0.0557	0.0557	0.0557	0.0557	0.0557	0.2542	0.5801
741	8	0.798	9.35	-0.00	-0.1666	-0.02	0.09	0.09	0.09	0.09	0.09	0.09	0.09	0.09	0.09	0.09	0.09	0.2354	0.5543
	8	0.0012	-0.0000	0.0003	-0.1666	1.5145	0.0757	0.0754	0.0757	0.0754	0.0757	0.0754	0.0757	0.0754	0.0757	0.0754	0.0757	0.2268	0.5492
	9	0.0001	0.0001	-0.0001	3.7855	-5.1224	0.0754	0.0754	0.0754	0.0754	0.0754	0.0754	0.0754	0.0754	0.0754	0.0754	0.0754	0.2268	0.5492
741	9	0.797	12.49	-0.00	-0.3076	-0.02	0.08	0.08	0.08	0.08	0.08	0.08	0.08	0.08	0.08	0.08	0.08	0.1668	0.5469
	8	0.0017	-0.0001	0.0003	-0.3076	0.9853	0.0856	0.0876	0.0856	0.0876	0.0856	0.0876	0.0856	0.0876	0.0856	0.0876	0.0856	0.1705	0.5386
	9	-0.0017	0.0002	-0.0000	-0.6689	0.0745	0.0876	0.0876	0.0876	0.0876	0.0876	0.0876	0.0876	0.0876	0.0876	0.0876	0.0876	0.1705	0.5386
741	10	0.797	-0.02	-0.00	0.3054	-0.02	-0.05	-0.05	-0.05	-0.05	-0.05	-0.05	-0.05	-0.05	-0.05	-0.05	-0.05	0.3219	0.4244
	8	0.0021	0.0001	0.0005	0.3054	1.3266	0.0026	0.0012	0.0026	0.0012	0.0026	0.0012	0.0026	0.0012	0.0026	0.0012	0.0026	0.4301	0.5480
	9	-0.0009	0.0002	-0.0001	-1.1509	0.5775	0.0012	0.0012	0.0012	0.0012	0.0012	0.0012	0.0012	0.0012	0.0012	0.0012	0.0012	0.4301	0.5480
742	1	0.798	-3.10	-0.00	-0.1419	-0.02	0.96	0.96	0.96	0.96	0.96	0.96	0.96	0.96	0.96	0.96	0.96	0.2751	0.6562
	8	0.0019	-0.0000	0.0003	-0.1419	0.8785	-0.0180	-0.0169	-0.0180	-0.0169	-0.0180	-0.0169	-0.0180	-0.0169	-0.0180	-0.0169	-0.0180	0.3434	0.6456
	9	-0.0018	0.0002	-0.0001	-0.5606	0.4999	-0.0169	-0.0169	-0.0169	-0.0169	-0.0169	-0.0169	-0.0169	-0.0169	-0.0169	-0.0169	-0.0169	0.3434	0.6456
742	2	0.796	-1.08	-0.00	-0.0617	-0.02	1.00	1.00	1.00	1.00	1.00	1.00	1.00	1.00	1.00	1.00	1.00	2.2226	-5.8559
	8	0.0025	-0.0000	0.0004	-0.0617	0.8106	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	1.1585	0.6986
	9	-0.0010	0.0002	-0.0001	-1.0473	0.6226	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	1.1585	0.6986
742	3	0.796	-0.03	-0.00	0.0098	-0.02	1.02	1.02	1.02	1.02	1.02	1.02	1.02	1.02	1.02	1.02	1.02	0.3423	0.6222
	8	0.0030	0.0000	0.0004	0.0098	0.8117	0.0070	0.0059	0.0070	0.0059	0.0070	0.0059	0.0070	0.0059	0.0070	0.0059	0.0070	0.3423	0.6222
	9	-0.0007	0.0002	-0.0001	-1.4218	0.7118	0.0059	0.0059	0.0059	0.0059	0.0059	0.0059	0.0059	0.0059	0.0059	0.0059	0.0059	0.3423	0.6222
742	4	0.796	0.51	-0.00	0.0124	-0.02	1.03	1.03	1.03	1.03	1.03	1.03	1.03	1.03	1.03	1.03	1.03	0.3048	0.6293
	8	0.0033	0.0000	0.0004	0.0124	0.7273	0.0121	0.0103	0.0121	0.0103	0.0121	0.0103	0.0121	0.0103	0.0121	0.0103	0.0121	0.3048	0.6293
	9	-0.0010	0.0002	-0.0001	-1.0436	0.5260	0.0103	0.0103	0.0103	0.0103	0.0103	0.0103	0.0103	0.0103	0.0103	0.0103	0.0103	0.3048	0.6293
742	5	0.798	1.00	-0.00	-0.0499	-0.02	1.04	1.04	1.04	1.04	1.04	1.04	1.04	1.04	1.04	1.04	1.04	0.2883	0.6234
	8	0.0038	-0.0000	0.0004	-0.0499	0.5751	0.0167	0.0142	0.0167	0.0142	0.0167	0.0142	0.0167	0.0142	0.0167	0.0142	0.0167	0.2883	0.6234
	9	-0.0012	0.0002	-0.0001	-0.8346	0.5151	0.0142	0.0142	0.0142	0.0142	0.0142	0.0142	0.0142	0.0142	0.0142	0.0142	0.0142	0.2883	0.6234
742	6	0.797	3.15	-0.00	0.0695	-0.02	1.08	1.08	1.08	1.08	1.08	1.08	1.08	1.08	1.08	1.08	1.08	0.2732	0.6232
	8	0.0028	0.0000	0.0004	0.0695	0.8284	0.0366	0.0337	0.0366	0.0337	0.0366	0.0337	0.0366	0.0337	0.0366	0.0337	0.0366	0.2732	0.6232
	9	-0.0010	0.0002	-0.0000	-1.1851	0.4675	0.0337	0.0337	0.0337	0.0337	0.0337	0.0337	0.0337	0.0337	0.0337	0.0337	0.0337	0.2732	0.6232
742	7	0.798	6.23	-0.00	-0.2219	-0.02	1.14	1.14	1.14	1.14	1.14	1.14	1.14	1.14	1.14	1.14	1.14	0.2790	0.5688
	8	0.0021	-0.0000	0.0003	-0.2219	0.7499	0.0580	0.0585	0.0580	0.0585	0.0580	0.0585	0.0580	0.0585	0.0580	0.0585	0.0580	0.2790	0.5688
	9	-0.0001	0.0002	-0.0000	-7.5579	-0.0361	0.0585	0.0585	0.0585	0.0585	0.0585	0.0585	0.0585	0.0585	0.0585	0.0585	0.0585	0.2790	0.5688

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																		
742	8	0.797	9.33	-0.002	-0.3121	-0.02	1.16	0.06	1.10	0.2274	0.5515	0.0026	-0.0001	0.0002	-0.5586	0.0794	0.0036	0.0087	0.2211	0.5433
	8	-0.0011	0.0002	0.0000	-1.2076	0.1207	0.0783	0.0034	0.0085											
	9																			
742	9	0.797	12.50	-0.003	-0.4121	-0.02	1.15	0.06	1.09	0.1551	0.5482	0.0021	-0.0001	0.0003	0.7535	0.0878	0.0027	0.0096	0.1614	0.5395
	9	-0.0015	0.0002	-0.0000	-0.6913	0.2548	0.0891	0.0028	0.0096											
	7																			
742	10	0.800	-0.01	-0.004	0.0170	-0.02	1.02	0.06	0.96	0.3169	0.6040	0.0029	0.0000	-0.0004	0.8312	0.0072	0.0004	0.0008	0.1461	0.6801
	8	-0.0008	0.0002	-0.0001	-1.2479	0.7441	0.0059	0.0001	0.0008											
	9																			
743	1	0.799	-3.08	-0.003	-0.1731	-0.02	2.96	-0.06	3.03	0.2699	0.7380	0.0022	-0.0002	-0.0003	0.7575	-0.0053	-0.0003	-0.0007	0.4695	0.6668
	8	-0.0022	0.0002	-0.0001	-0.5773	0.2468	-0.0053	-0.0003	-0.0007											
	9																			
743	2	0.799	-1.03	-0.003	-0.1550	-0.02	3.00	0.06	3.06	0.3298	0.6534	0.0030	-0.0003	-0.0003	0.6389	0.0095	0.0006	0.0012	0.1957	0.6694
	8	-0.0030	0.0000	-0.0003	-1.0449	0.6466	0.0095	0.0003	0.0012											
	9	-0.0010	0.0002	-0.0001	-1.6572	1.7291	0.0176	0.0007	0.0022											
743	3	0.798	-0.02	-0.005	0.2019	-0.02	3.02	0.06	3.08	0.2940	0.6429	0.0023	-0.0001	-0.0005	0.7440	0.0184	0.0010	0.0023	0.2262	0.6447
	8	-0.0023	0.0000	-0.0005	-1.6572	1.7291	0.0176	0.0007	0.0022											
	9	-0.0005	0.0001	-0.0001	-1.7590	1.6277	0.0219	0.0010	0.0028											
743	4	0.799	0.50	-0.004	0.0090	-0.02	3.03	0.06	3.08	0.2865	0.6410	0.0033	-0.0001	-0.0004	0.7509	0.0232	0.0013	0.0028	0.2446	0.6503
	8	-0.0033	0.0000	-0.0004	-0.9093	0.4146	0.0263	0.0010	0.0028											
	9	-0.0005	0.0001	-0.0001	-1.7590	1.6277	0.0219	0.0010	0.0028											
743	5	0.799	1.04	-0.004	-0.0312	-0.02	3.04	0.06	3.10	0.2794	0.6334	0.0037	-0.0002	-0.0004	0.6206	0.0285	0.0015	0.0034	0.2512	0.6460
	8	-0.0037	0.0000	-0.0004	-0.9093	0.4146	0.0263	0.0010	0.0028											
	9	-0.0012	0.0002	-0.0001	-1.4555	0.2216	0.0463	0.0022	0.0057											
743	6	0.799	3.14	-0.004	0.0047	-0.02	3.08	0.06	3.13	0.2723	0.6090	0.0034	-0.0002	-0.0004	0.6772	0.0477	0.0026	0.0057	0.2440	0.6172
	8	-0.0034	0.0000	-0.0004	-1.1455	0.2216	0.0463	0.0022	0.0057											
	9	-0.0011	0.0002	-0.0000	-1.4555	0.2216	0.0463	0.0022	0.0057											
743	7	0.799	6.24	-0.004	0.1329	-0.02	3.12	0.06	3.17	0.2456	0.5638	0.0013	-0.0002	-0.0004	1.5259	0.0675	0.0033	0.0075	0.2417	0.5597
	8	-0.0013	0.0000	-0.0004	4.3359	-0.7097	0.0673	0.0032	0.0075											
	9	0.0002	0.0002	-0.0000	4.3359	-0.7097	0.0673	0.0032	0.0075											
743	8	0.783	9.35	-0.003	-0.3051	-0.02	3.13	0.06	3.18	0.1985	0.5443	0.0022	-0.0003	-0.0003	0.7220	0.0826	0.0032	0.0090	0.1955	0.5598
	8	-0.0022	-0.0001	0.0003	-0.9366	-0.2122	0.0841	0.0032	0.0090											
	9	-0.0016	0.0003	0.0000	-0.9366	-0.2122	0.0841	0.0032	0.0090											

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key														
743	9	0.799	-0.0001	12.50	-0.0001	-0.0002	-0.4793	0.7131	3.13	0.06	3.18	0.0098	0.1458	0.5486		
	8	-0.0015	0.0000	-0.0001	0.0002	0.3578	0.0910	0.0027	0.0094	0.0026	0.0097	0.0097	0.1531	0.5343		
743	10	0.799	0.0000	0.01	0.0005	-0.0001	0.0464	0.9091	3.02	0.06	3.07	0.0023	0.2828	0.6263		
	8	-0.0009	0.0002	-0.0001	-0.0001	0.6200	0.0175	0.0007	0.0187	0.0010	0.0022	0.0022	0.2828	0.6357		
744	1	0.800	-3.05	0.0000	-0.0003	-0.02	6.01	0.06	6.01	0.06	6.01	0.0010	0.3408	0.6508		
	8	0.0020	0.0000	0.0001	0.0001	0.8935	0.0080	0.0003	0.0079	0.0005	0.0011	0.0011	0.2158	0.7270		
	9	-0.0016	0.0001	-0.0001	-0.0001	0.5729	0.0080	0.0003	0.0080	0.0003	0.0011	0.0011	0.2158	0.7270		
744	3	0.799	-1.01	0.0000	-0.0004	-0.02	6.05	0.06	6.05	0.06	6.05	0.0034	0.2824	0.6439		
	8	0.0024	0.0002	0.0000	0.0004	0.9253	0.0270	0.0015	0.0270	0.0015	0.0034	0.0034	0.2475	0.6527		
	9	-0.0004	0.0002	-0.0002	-0.0001	1.6053	0.0265	0.0013	0.0265	0.0013	0.0034	0.0034	0.2475	0.6527		
744	4	0.799	-0.0000	0.0000	-0.0004	-0.02	6.07	0.06	6.07	0.06	6.07	0.0046	0.2723	0.6359		
	8	0.0032	0.0000	0.0000	0.0004	0.7145	0.0367	0.0020	0.0367	0.0020	0.0046	0.0046	0.2440	0.6415		
	9	-0.0010	0.0002	-0.0001	-0.0001	0.6645	0.0356	0.0017	0.0356	0.0017	0.0045	0.0045	0.2440	0.6415		
744	5	0.799	0.50	0.0000	-0.0001	-0.02	6.08	0.06	6.08	0.06	6.07	0.0052	0.2716	0.6278		
	8	0.0032	0.0000	0.0000	0.0004	0.7555	0.0417	0.0022	0.0417	0.0022	0.0052	0.0052	0.2443	0.6345		
	9	-0.0010	0.0002	-0.0001	-0.0001	0.5983	0.0405	0.0019	0.0405	0.0019	0.0051	0.0051	0.2443	0.6345		
744	6	0.800	1.05	0.0000	-0.0001	-0.02	6.09	0.06	6.09	0.06	6.08	0.0056	0.2707	0.6148		
	8	0.0033	0.0000	0.0000	0.0005	0.7488	0.0461	0.0021	0.0461	0.0021	0.0056	0.0056	0.2444	0.6300		
	9	-0.0008	0.0002	-0.0001	-0.0001	0.9342	0.0446	0.0021	0.0446	0.0021	0.0056	0.0056	0.2444	0.6300		
744	7	0.799	3.13	0.0000	-0.0002	-0.02	6.12	0.06	6.12	0.06	6.11	0.0067	0.2637	0.5717		
	8	0.0031	0.0000	0.0000	0.0004	0.7116	0.0593	0.0029	0.0593	0.0029	0.0067	0.0067	0.2557	0.5813		
	9	0.0006	0.0001	-0.0001	-0.0002	-2.1045	0.0583	0.0029	0.0583	0.0029	0.0067	0.0067	0.2557	0.5813		
744	8	0.799	6.26	0.0000	-0.0003	-0.02	6.14	0.06	6.14	0.06	6.14	0.0088	0.2207	0.5548		
	8	0.0020	0.0001	0.0000	0.0000	0.9554	0.0790	0.0034	0.0790	0.0034	0.0088	0.0088	0.2185	0.5468		
	9	-0.0004	0.0003	0.0000	0.0000	-0.3865	0.0790	0.0034	0.0790	0.0034	0.0088	0.0088	0.2185	0.5468		
744	9	0.799	9.34	0.0000	-0.0003	-0.02	6.13	0.06	6.13	0.06	6.13	0.0093	0.1547	0.5497		
	8	0.0016	0.0001	0.0000	0.0000	1.0115	0.0853	0.0026	0.0853	0.0026	0.0093	0.0093	0.1602	0.5441		
	9	0.0001	0.0001	-0.0001	-0.0001	-5.1889	0.0878	0.0028	0.0878	0.0028	0.0095	0.0095	0.1602	0.5441		
744	10	0.800	12.53	0.0000	-0.0002	-0.02	6.14	0.06	6.14	0.06	6.13	0.0101	0.1389	0.5475		
	8	0.0021	0.0002	0.0000	0.0002	0.6939	0.0923	0.0025	0.0923	0.0025	0.0101	0.0101	0.1385	0.5375		
	9	-0.0018	0.0001	-0.0001	-0.0001	0.3221	0.0940	0.0026	0.0940	0.0026	0.0101	0.0101	0.1385	0.5375		

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd																		
744	11	7	0.800	0.01	-0.00	-0.02	6.07	0.06	6.06	0.2669	0.6268									
		8	0.0039	-0.0001	0.0004	0.5166	0.370	0.0019	0.0046	0.2446	0.6423									
		9	-0.0016	0.0002	-0.0000	0.1504	0.0356	0.0017	0.0045											
745	1	7	0.801	-3.04	-0.00	-0.02	9.08	0.06	9.03	0.2867	0.6568									
		8	0.0019	-0.0000	0.0003	0.9407	0.248	0.0014	0.0032	0.2456	0.6657									
		9	-0.0010	0.0001	-0.0002	1.1458	0.0251	0.0012	0.0033											
745	2	7	0.800	-1.01	-0.00	-0.02	9.12	0.06	9.06	0.2717	0.6261									
		8	0.0031	-0.0001	0.0003	0.5720	0.445	0.0024	0.0055	0.2413	0.6319									
		9	-0.0015	0.0002	-0.0000	0.2530	0.0445	0.0021	0.0056											
745	3	7	0.800	0.02	-0.00	-0.02	9.14	0.05	9.08	0.2791	0.6021									
		8	0.0028	0.0000	0.0004	0.8826	0.0520	0.0029	0.0062	0.2477	0.6106									
		9	-0.0007	0.0001	-0.0001	1.1923	0.0513	0.0025	0.0062											
745	4	7	0.800	0.52	-0.00	-0.02	9.15	0.06	9.09	0.2836	0.5853									
		8	0.0032	0.0000	0.0005	0.7893	0.0545	0.0030	0.0063	0.2485	0.5921									
		9	-0.0001	0.0001	-0.0002	10.2432	0.0546	0.0027	0.0064											
745	5	7	0.800	1.08	-0.00	-0.02	9.15	0.06	9.09	0.2664	0.5786									
		8	0.0035	0.0000	0.0004	0.6775	0.0576	0.0030	0.0066	0.2516	0.5872									
		9	-0.0014	0.0002	-0.0000	0.3432	0.0571	0.0028	0.0067											
745	6	7	0.800	3.17	-0.00	-0.02	9.18	0.06	9.11	0.2307	0.5659									
		8	0.0034	-0.0000	0.0004	0.6262	0.736	0.0034	0.0081	0.1818	0.5493									
		9	-0.0014	0.0002	-0.0000	0.2229	0.0722	0.0032	0.0081	0.1839	0.5444									
745	7	7	0.800	6.26	-0.00	-0.02	9.18	0.06	9.11	0.1414	0.5541									
		8	0.0020	-0.0001	0.0003	0.8462	0.0833	0.0030	0.0091	0.1240	0.5532									
		9	0.0002	0.0002	-0.0000	0.4569	0.0847	0.0031	0.0092	0.1477	0.5429									
745	8	7	0.801	9.36	-0.00	-0.02	9.17	0.06	9.10	0.1240	0.5532									
		8	0.0019	-0.0001	0.0003	0.8092	0.0888	0.0025	0.0098	0.1229	0.5492									
		9	-0.0001	0.0001	-0.0001	5.9529	0.0901	0.0026	0.0097											
745	9	7	0.800	12.54	-0.00	-0.02	9.18	0.06	9.10	0.2775	0.6003									
		8	0.0018	-0.0002	0.0003	0.8610	0.0967	0.0024	0.0062	0.2474	0.6108									
		9	-0.0017	0.0002	-0.0001	0.3190	0.0978	0.0024	0.0062											
745	10	7	0.800	0.03	-0.00	-0.02	9.15	0.06	9.07	0.2775	0.6003									
		8	0.0031	-0.0000	0.0004	0.7428	0.0521	0.0028	0.0062											
		9	-0.0012	0.0002	-0.0001	0.4707	0.0515	0.0025	0.0062											

See Key

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key												
746	1	7	0.801	-3.01	-0.0001	-0.0003	-0.2523	-0.7371	-0.02	12.07	0.06	12.08	0.2694	0.6347
		6	0.0020	-0.0001	0.0001	0.0002	-0.7371	1.1606	0.7492	0.0428	0.0023	0.0054	0.2463	0.6455
		9	-0.0010	0.0000	-0.0002	-0.0002	-0.7371	1.1606	1.1606	0.0436	0.0021	0.0056	0.2463	0.6455
746	2	7	0.800	-0.99	-0.0002	-0.0004	-0.1171	-1.3147	-0.02	12.10	0.06	12.11	0.2698	0.5043
		8	0.0028	-0.0000	0.0000	0.0004	-0.1171	0.7323	0.7323	0.0560	0.0030	0.0065	0.2496	0.5910
		9	-0.0008	0.0002	-0.0001	-0.0001	-1.3147	0.8295	0.8295	0.0568	0.0028	0.0067	0.2496	0.5910
746	3	7	0.798	0.02	-0.0002	-0.0005	0.0701	-1.0358	-0.02	12.11	0.06	12.11	0.2457	0.5701
		8	0.0026	0.0000	0.0000	0.0005	0.0701	0.4824	0.9423	0.0642	0.0031	0.0074	0.2348	0.5798
		9	-0.0011	0.0002	-0.0001	-0.0001	-1.0358	0.4824	0.4824	0.0638	0.0029	0.0073	0.2348	0.5798
746	4	7	0.799	0.53	-0.0002	-0.0005	0.0635	-1.1959	-0.01	12.12	0.06	12.12	0.2382	0.5763
		8	0.0031	0.0000	0.0000	0.0005	0.0635	0.8198	0.8198	0.0684	0.0032	0.0078	0.2295	0.5763
		9	-0.0008	0.0002	-0.0001	-0.0001	-1.1959	0.9712	0.9712	0.0675	0.0030	0.0077	0.2295	0.5763
746	5	7	0.799	1.11	-0.0002	-0.0004	-0.0820	-0.8388	-0.02	12.12	0.06	12.13	0.2317	0.5713
		8	0.0039	-0.0000	0.0000	0.0004	-0.0820	0.5761	0.5761	0.0719	0.0033	0.0082	0.2317	0.5713
		9	-0.0014	0.0002	-0.0000	-0.0000	-0.8388	0.3347	0.3347	0.0718	0.0032	0.0081	0.2317	0.5713
746	6	7	0.798	3.18	-0.0002	-0.0004	-0.0671	-1.7411	-0.02	12.14	0.06	12.14	0.2095	0.5486
		8	0.0032	-0.0000	0.0000	0.0004	-0.0671	0.6590	0.6590	0.0816	0.0034	0.0089	0.2095	0.5486
		9	-0.0006	0.0002	-0.0001	-0.0001	-1.7411	1.2450	1.2450	0.0817	0.0034	0.0089	0.2095	0.5486
746	7	7	0.800	6.26	-0.0003	-0.0005	-0.2732	-14.3264	-0.02	12.12	0.06	12.13	0.1496	0.5546
		8	0.0021	-0.0001	0.0000	0.0003	-0.2732	0.7900	0.7900	0.0854	0.0025	0.0094	0.1496	0.5546
		9	-0.0000	0.0003	-0.0000	-0.0000	-14.3264	23.0359	23.0359	0.0873	0.0026	0.0095	0.1496	0.5546
746	8	7	0.799	9.33	-0.0002	-0.0003	-0.3549	-1.3531	-0.02	12.13	0.06	12.13	0.1392	0.5506
		8	0.0018	-0.0001	0.0000	0.0003	-0.3549	0.9495	0.9495	0.0923	0.0025	0.0101	0.1392	0.5506
		9	-0.0008	0.0002	-0.0000	-0.0000	-1.3531	0.2777	0.2777	0.0939	0.0026	0.0101	0.1392	0.5506
746	9	7	0.800	12.54	-0.0003	-0.0002	-0.6844	-0.4680	-0.02	12.13	0.06	12.12	0.1182	0.5521
		8	0.0023	-0.0003	0.0000	0.0002	-0.6844	0.6369	0.6369	0.0997	0.0023	0.0110	0.1182	0.5521
		9	-0.0025	0.0002	-0.0000	-0.0000	-0.4680	0.1577	0.1577	0.1006	0.0023	0.0111	0.1182	0.5521
746	10	7	0.800	0.07	-0.0002	-0.0005	0.1119	-1.4318	-0.02	12.11	0.06	12.12	0.2427	0.5741
		8	0.0025	0.0000	0.0000	0.0005	0.1119	1.0663	1.0663	0.0631	0.0029	0.0073	0.2427	0.5741
		9	-0.0007	0.0002	-0.0001	-0.0001	-1.4318	1.1974	1.1974	0.0639	0.0029	0.0073	0.2427	0.5741
747	1	7	0.799	-2.98	-0.0001	-0.0003	-0.0078	-0.8213	-0.02	15.11	0.06	15.10	0.2700	0.5881
		8	0.0014	-0.0000	0.0000	0.0003	-0.0078	1.2954	1.2954	0.0549	0.0029	0.0064	0.2700	0.5881
		9	-0.0010	0.0001	-0.0002	-0.0002	-0.8213	1.0931	1.0931	0.0565	0.0027	0.0067	0.2700	0.5881

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pos.	α	See Key												
747	2	7 8 9	0.800 0.0033 -0.0014	-0.0001 0.0002	-0.0003 0.0001	-0.2583 -0.8383	-0.02 0.5222 0.3532	15.14 0.709 0.0718	0.06 0.0032 0.0031	15.12 0.0081 0.0082	0.2308 0.2204	0.5746 0.5755			
747	3	7 8 9	0.799 0.0027 -0.0010	0.0002 0.0002	-0.0001 0.0001	0.0607 -1.0370	-0.01 0.9460 0.5970	15.15 0.767 0.0776	0.06 0.0034 0.0033	15.13 0.0087 0.0088	0.2252 0.2137	0.5691 0.5670			
747	4	7 8 9	0.799 0.0033 -0.0006	0.0000 0.0001	-0.0004 0.0001	0.0027 -1.4130	-0.02 0.7348 1.3405	15.15 0.795 0.0796	0.06 0.0034 0.0033	15.13 0.0089 0.0088	0.2193 0.2079	0.5516 0.5583			
747	5	7 8 9	0.801 0.0034 -0.0005	1.09 0.0001	-0.0005 0.0001	0.0446 -1.7743	-0.01 0.7561 1.9554	15.15 0.811 0.0811	0.06 0.0034 0.0033	15.13 0.0089 0.0089	0.2124 0.2065	0.5573 0.5518			
747	6	7 8 9	0.799 0.0029 -0.0016	3.18 0.0003	-0.0004 0.0000	0.0288 -0.9253	-0.02 0.8375 0.0683	15.15 0.846 0.0861	0.06 0.0029 0.0030	15.13 0.0093 0.0094	0.1723 0.1763	0.5522 0.5483			
747	7	7 8 9	0.800 0.0021 -0.0002	6.27 0.0003	-0.0003 0.0000	-0.2292 -6.3548	-0.02 0.8605 -1.1413	15.14 0.892 0.0906	0.06 0.0024 0.0026	15.12 0.0099 0.0100	0.1396 0.1445	0.5573 0.5533			
747	8	7 8 9	0.799 0.0023 -0.0014	9.37 0.0002	-0.0003 0.0000	-0.4805 -0.8424	-0.02 0.6632 0.1757	15.15 0.971 0.0981	0.06 0.0024 0.0024	15.12 0.0108 0.0108	0.1247 0.1248	0.5561 0.5535			
747	9	7 8 9	0.798 0.0016 -0.0011	12.54 0.0001	-0.0004 0.0002	-0.5368 -0.5968	-0.03 1.2766 1.0172	15.15 0.1042 0.1052	0.06 0.0023 0.0022	15.12 0.0115 0.0116	0.1123 0.1088	0.5562 0.5523			
747	10	7 8 9	0.799 0.0028 -0.0006	0.05 0.0002	-0.0005 0.0001	0.0518 -1.5370	-0.02 0.9218 1.2608	15.15 0.0770 0.0778	0.06 0.0034 0.0033	15.12 0.0087 0.0087	0.2227 0.2129	0.5659 0.5652			
748	5	7 8 9	0.800 0.0109 -0.0114	3.08 0.0009	-0.0013 0.0013	0.4129 0.3945	-3.03 0.6115 0.5735	2.95 -0.0089 -0.0377	2.95 -0.0023 -0.0023	3.04 -0.0011 -0.0046	0.2024 0.3064	0.6614 0.6227			
748	6	7 8 9	0.799 0.0095 -0.0120	-0.0009 0.0009	-0.0011 0.0014	0.4383 0.4019	-3.01 0.6190 0.6021	3.01 0.0162 -0.0110	2.95 0.0010 -0.0008	-2.98 0.0020 -0.0015	0.3165 0.3904	0.6405 0.6800			

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd																						
748	7	0.798	-0.0090	0.0111	0.0010	0.0100	0.4840	-3.501	0.6504	3.01	0.169	2.95	-2.98	0.3169	0.6469								
	8	-0.0111	0.0010	-0.0015	0.0015	-0.0011	0.4614	0.6852	-0.0106	-0.0008	0.0010	-0.0014	0.3938	0.6834									
	9	0.798	-0.0086	0.0115	0.0009	0.0114	0.5294	-2.99	0.6938	3.03	0.258	2.94	-2.96	0.2981	0.6406								
	8	-0.0115	0.0009	-0.0014	0.0014	-0.0011	0.4273	0.6309	-0.0036	-0.0004	0.0015	-0.0005	0.5845	0.7645									
	9	0.800	-0.0095	0.0123	0.0009	0.0113	0.3728	-2.99	0.5601	3.05	0.363	2.95	-2.95	0.2808	0.6298								
	8	-0.0123	0.0009	-0.0013	0.0013	-0.0010	0.3811	0.5560	0.0034	-0.0000	0.0020	0.0045	-0.0153	0.5919									
	9	0.801	-0.0101	0.0114	0.0009	0.0111	0.3590	-3.01	0.5447	3.08	0.454	2.95	-2.93	0.2790	0.6134								
	8	-0.0114	0.0009	-0.0014	0.0014	-0.0011	0.4351	0.6291	0.0123	0.0004	0.0025	0.0014	0.1771	0.6061									
	9	0.799	-0.0112	0.0117	0.0010	0.0115	0.3134	-2.99	0.5143	3.09	0.527	2.95	-2.91	0.2838	0.5921								
	8	-0.0117	0.0010	-0.0015	0.0015	-0.0011	0.4479	0.6471	0.0217	0.0009	0.0029	0.0025	0.2207	0.5978									
	9	0.800	-0.0093	0.0117	0.0010	0.0113	0.4606	-2.99	0.6309	3.01	0.164	2.95	-2.98	0.3885	0.6293								
	8	-0.0117	0.0010	-0.0015	0.0015	-0.0011	0.4365	0.6415	-0.0109	-0.0008	0.0010	-0.0015	0.3956	0.6293									
	9	0.799	-0.0122	0.0119	0.0010	0.0115	0.4187	-3.00	0.6288	-0.0214	0.12	2.95	-0.08	0.3588	0.6369								
	8	-0.0119	0.0010	-0.0015	0.0015	-0.0011	0.4334	0.6483	-0.0214	-0.0014	0.0013	-0.0027	0.3270	0.6386									
	9	0.800	-0.0113	0.0133	0.0010	0.0112	0.3397	-2.99	0.5429	0.06	0.013	2.95	-0.03	0.6235	0.3542								
	8	-0.0133	0.0010	-0.0015	0.0015	-0.0011	0.3853	0.5859	0.0023	-0.0000	0.0001	0.0003	-0.0643	0.7114									
	9	0.800	-0.0094	0.0121	0.0011	0.0117	0.5358	-3.00	0.7271	0.05	0.048	2.95	-0.01	0.4091	0.5980								
	8	-0.0121	0.0011	-0.0017	0.0017	-0.0013	0.4693	0.7147	0.0059	0.0001	0.0001	0.0008	0.1571	0.7131									
	9	0.798	-0.0110	0.0128	0.0010	0.0112	0.3557	-2.99	0.5664	0.04	0.092	2.95	-0.01	0.3540	0.6271								
	8	-0.0128	0.0010	-0.0016	0.0016	-0.0012	0.4215	0.6353	0.0104	0.0004	0.0006	0.0013	0.1930	0.6641									
	9	0.800	-0.0109	0.0138	0.0010	0.0112	0.3498	-2.99	0.5671	0.02	0.182	2.95	-0.00	0.3009	0.6241								
	8	-0.0138	0.0010	-0.0015	0.0015	-0.0011	0.3731	0.5561	0.0186	0.0008	0.0010	0.0023	0.2310	0.6382									

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Pt. Cd	See Key																								
749	6	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	
		0.798	-0.0110	0.0130	3.10	-0.0008	0.0011	-0.0016	-0.0013	0.3722	0.4301	-2.99	0.5922	0.6481	0.0285	0.0016	2.95	0.0013	0.0035	0.0034	0.0035	0.2096	0.2430	0.6253	0.6261	
749	7	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	7
		0.798	-0.0114	0.0120	4.15	-0.0008	0.0012	-0.0019	-0.0013	0.3822	0.5295	-2.99	0.6032	0.7958	0.0374	0.0019	2.95	0.0020	0.0046	0.0046	0.0046	0.2789	0.2459	0.6205	0.6251	
749	8	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	7
		0.799	-0.0103	0.0127	-0.0008	0.0010	-0.0016	-0.0013	0.4341	0.4291	-2.99	0.6314	0.6477	0.0021	-0.0000	2.95	0.0001	0.0000	0.0002	0.0000	0.0002	0.5274	-0.1308	0.2117	0.5807	
750	1	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	7
		0.798	-0.0048	-0.0043	-3.10	0.0000	-0.0004	-0.0005	-0.0012	-0.0199	-0.0413	0.48	0.5657	0.5083	-0.0244	-0.0014	-0.48	-0.0012	-0.0031	-0.0029	-0.0029	0.2626	0.3241	0.6343	0.6434	
750	2	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	7
		0.799	-0.0054	-0.0028	-0.0001	0.0000	-0.0004	-0.0007	0.1258	0.1408	0.49	0.7051	0.8742	0.0021	-0.0000	-0.48	-0.0002	0.0000	0.0002	0.0001	0.0001	0.4980	-0.6617	0.5089	0.8300	
750	3	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	7
		0.799	-0.0051	-0.0019	0.49	0.0002	-0.0005	-0.0008	0.2397	0.1785	0.49	0.8221	1.4861	0.0048	-0.0001	-0.48	-0.0004	0.0001	0.0006	0.0006	0.0006	0.3769	0.1321	0.6229	0.6753	
750	4	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	7
		0.798	-0.0064	-0.0030	1.00	0.0000	-0.0004	-0.0007	0.0572	0.0280	0.49	0.5995	0.7653	0.0067	-0.0003	-0.48	-0.0006	0.0011	0.0013	0.0011	0.0011	0.3245	0.1750	0.6152	0.6370	
750	5	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	7
		0.798	-0.0062	-0.0027	2.05	0.0001	-0.0005	-0.0007	0.0930	0.0481	0.49	0.6214	0.9359	0.0174	-0.0007	-0.47	-0.0017	0.0021	0.0024	0.0021	0.0021	0.2917	0.2235	0.6209	0.6247	
750	6	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	7
		0.799	-0.0056	-0.0034	3.11	0.0001	-0.0004	-0.0007	0.1479	-0.0742	0.49	0.6618	0.5868	0.0270	-0.0013	-0.47	-0.0016	0.0035	0.0035	0.0035	0.0035	0.2829	0.2426	0.6221	0.6181	
750	7	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	7
		0.799	-0.0053	-0.0025	4.15	0.0001	-0.0004	-0.0006	0.0936	-0.0839	0.49	0.6390	0.8040	0.0374	-0.0018	-0.48	-0.0021	0.0046	0.0046	0.0046	0.0046	0.2742	0.2469	0.6199	0.6219	
750	8	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	7	8	9	7
		0.798	-0.0049	-0.0027	-0.0002	0.0000	-0.0004	-0.0008	0.2631	0.0087	0.49	0.8659	0.8500	0.0012	-0.0001	-0.47	-0.0002	0.0001	0.0001	0.0001	0.0001	0.4141	-0.4049	0.4924	0.4894	

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key													
751	1	7 8 9	0.797 0.0041 -0.0036	-3.10 0.0000 0.0000	-0.00 0.0006 -0.0004	0.0987 -0.0106 0.0000	0.49 0.7408 0.6598	-0.57 -0.0272 -0.0204	-0.47 -0.0014 -0.0013	0.42 -0.0034 -0.0026	0.2615 0.5274 0.6316	0.6404				
751	2	7 8 9	0.798 0.0055 -0.0028	-0.04 0.0001 0.0000	-0.00 0.0007 -0.0004	0.1224 -0.0310 0.0000	0.49 0.6963 0.7765	-0.50 0.0004 0.0031	-0.47 0.0001 0.0000	0.47 -0.0000 -0.0004	1.1717 -0.0657 -0.1296	0.6228				
751	3	7 8 9	0.797 0.0059 -0.0036	0.51 0.0001 0.0000	-0.00 0.0007 -0.0003	0.0921 -0.1151 0.0000	0.49 0.6328 0.4847	-0.50 0.0039 0.0068	-0.47 0.0003 0.0002	0.47 0.0004 0.0009	0.4127 0.1754 0.5709	0.6776				
751	4	7 8 9	0.798 0.0059 -0.0032	1.02 0.0001 0.0000	-0.00 0.0007 -0.0004	0.1033 -0.0593 0.0000	0.49 0.6554 0.6358	-0.48 0.0082 0.0111	-0.47 0.0005 0.0004	0.48 0.0010 0.0014	0.3409 0.2047 0.6297	0.6485				
751	5	7 8 9	0.797 0.0057 -0.0023	2.05 0.0001 -0.0000	-0.00 0.0008 -0.0005	0.1629 0.0838 0.0000	0.49 0.7135 1.1330	-0.47 0.0172 0.0201	-0.47 0.0010 0.0009	0.50 0.0021 0.0025	0.2978 0.2283 0.6314	0.6218				
751	6	7 8 9	0.799 0.0049 -0.0035	3.10 0.0002 0.0000	-0.00 0.0007 -0.0003	0.2338 -0.1148 0.0000	0.49 0.7962 0.5187	-0.44 0.0267 0.0299	-0.47 0.0015 0.0014	0.52 0.0033 0.0036	0.2859 0.2427 0.6174					
751	7	7 8 9	0.797 0.0050 -0.0026	4.14 0.0001 0.0000	-0.00 0.0006 -0.0003	0.1234 -0.1240 0.0000	0.49 0.6856 0.7264	-0.42 0.0364 0.0405	-0.47 0.0020 0.0019	0.54 0.0045 0.0049	0.2760 0.2411 0.6182	0.6138				
751	8	7 8 9	0.798 0.0053 -0.0029	-0.07 0.0001 0.0000	-0.00 0.0007 -0.0004	0.1431 -0.0457 0.0000	0.49 0.7249 0.7641	-0.51 0.0003 0.0028	-0.47 0.0000 0.0000	0.46 -0.0000 -0.0003	0.9701 -0.0038 -1.1482	0.6136				
752	1	7 8 9	0.798 0.0077 -0.0071	-3.10 0.0000 -0.0000	-0.00 0.0009 -0.0007	0.0938 0.0507 0.0000	1.02 0.5867 0.4917	-0.11 -0.0241 -0.0232	-1.02 -0.0012 -0.0014	0.07 -0.0030 -0.0029	0.3680 0.3205 0.6363	0.6408				
752	2	7 8 9	0.797 0.0088 -0.0059	-0.06 0.0002 -0.0001	-0.00 0.0010 -0.0007	0.1148 0.1150 0.0000	1.02 0.6001 0.6610	-0.04 0.0026 0.0007	-1.02 0.0002 -0.0001	0.02 -0.0001 -0.0001	0.4303 -0.6719 0.5218	0.6981				
752	3	7 8 9	0.798 0.0087 -0.0056	0.49 0.0003 -0.0001	-0.00 0.0011 -0.0007	0.1733 0.1306 0.0000	1.02 0.6382 0.6902	-0.03 0.0064 0.0044	-1.02 0.0004 0.0001	0.01 -0.0007 -0.0006	0.3520 0.1338 0.6141	0.7006				

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																						
752	4	7	0.798	1.02	0.1309	0.5822	0.6986	-0.02	-1.02	0.0013	0.3161	0.6162	0.0092	0.0002	0.0010	0.1482	0.0988	0.5416	0.5038	0.0200	0.0011	0.1728	0.6377	
752	5	8	0.0056	-0.0001	-0.0007	-0.0010	-0.0007	0.0110	0.0002	0.0010	0.0002	0.0011	0.0013	0.0092	0.0002	0.0010	0.1482	0.0988	0.5416	0.5038	0.0200	0.0011	0.1728	0.6377
752	5	9	0.798	2.05	0.0988	0.5416	0.5038	-0.00	-1.02	0.0025	0.2917	0.6281	0.0098	0.0001	0.0010	0.0547	0.0988	0.5416	0.5038	0.0200	0.0021	0.2204	0.6300	
752	6	7	0.799	3.10	0.2254	0.6703	0.5027	0.01	-1.02	0.0037	0.2831	0.6238	0.0083	0.0012	0.0006	0.2254	0.0368	0.6703	0.5027	0.0268	0.0032	0.2393	0.6140	
752	6	8	0.0063	-0.0000	-0.0006	-0.0011	-0.0006	0.03	-1.02	0.0048	0.2736	0.6199	0.0063	0.0016	0.0006	0.0368	0.0634	0.6149	0.0372	0.04	0.0046	0.2428	0.6194	
752	7	7	0.798	4.16	0.0988	0.5461	0.6149	0.04	-1.02	0.0048	0.2736	0.6199	0.0087	0.0021	0.0009	0.0988	0.0634	0.6149	0.0372	0.04	0.0046	0.2428	0.6194	
752	8	7	0.0086	-0.0003	-0.0006	-0.0011	-0.0006	0.04	-1.02	0.0048	0.2736	0.6199	0.0086	0.0021	0.0009	0.1848	0.0440	0.6591	0.5115	0.0028	0.0001	0.4066	0.5206	
752	8	8	0.799	5.04	0.1848	0.6591	0.5115	0.04	-1.02	0.0048	0.2736	0.6199	0.0086	0.0021	0.0009	0.1848	0.0440	0.6591	0.5115	0.0028	0.0001	0.4066	0.5206	
753	1	7	0.799	3.11	0.1203	0.6430	0.6989	-0.01	-1.01	0.0037	0.2711	0.6443	0.0070	0.0015	0.0009	0.1203	0.0909	0.6430	0.6989	0.0290	0.0022	0.2711	0.6443	
753	2	7	0.0055	-0.0001	-0.0007	-0.0007	-0.0007	0.01	-1.01	0.0037	0.2711	0.6443	0.0055	0.0011	0.0007	0.1203	0.0909	0.6430	0.6989	0.0290	0.0022	0.2711	0.6443	
753	2	8	0.799	4.02	0.1624	0.6474	0.7792	-0.01	-1.01	0.0037	0.2711	0.6443	0.0079	0.0010	0.0007	0.1624	0.1443	0.6474	0.7792	0.0052	0.0007	0.1834	0.6902	
753	3	7	0.0050	-0.0001	-0.0007	-0.0007	-0.0007	0.01	-1.01	0.0037	0.2711	0.6443	0.0050	0.0010	0.0007	0.1624	0.1443	0.6474	0.7792	0.0052	0.0007	0.1834	0.6902	
753	3	8	0.798	0.50	0.2016	0.6727	0.7481	-0.03	-1.01	0.0026	0.4113	0.5061	0.0084	0.0003	0.0007	0.2016	0.1183	0.6727	0.7481	0.0026	0.0012	0.4113	0.5061	
753	4	7	0.0054	-0.0001	-0.0007	-0.0007	-0.0007	0.01	-1.01	0.0026	0.4113	0.5061	0.0054	0.0003	0.0007	0.2016	0.1183	0.6727	0.7481	0.0026	0.0012	0.4113	0.5061	
753	4	8	0.798	1.02	0.2573	0.7429	0.7146	-0.01	-1.01	0.0026	0.3516	0.6205	0.0077	0.0004	0.0007	0.2573	0.1128	0.7429	0.7146	0.0062	0.0017	0.2142	0.6453	
753	5	7	0.0052	-0.0001	-0.0007	-0.0007	-0.0007	0.01	-1.01	0.0026	0.3516	0.6205	0.0052	0.0004	0.0007	0.2573	0.1128	0.7429	0.7146	0.0062	0.0017	0.2142	0.6453	
753	5	8	0.798	2.06	0.2442	0.7092	0.6582	-0.01	-1.01	0.0026	0.2942	0.6147	0.0078	0.0009	0.0007	0.2442	0.0926	0.7092	0.6582	0.0025	0.0028	0.2942	0.6147	
753	6	7	0.0054	-0.0001	-0.0007	-0.0007	-0.0007	0.01	-1.01	0.0026	0.2942	0.6147	0.0054	0.0009	0.0007	0.2442	0.0926	0.7092	0.6582	0.0025	0.0028	0.2942	0.6147	
753	6	8	0.799	3.09	0.1792	0.6374	0.6457	-0.01	-1.01	0.0026	0.2876	0.6200	0.0080	0.0014	0.0006	0.1792	0.1770	0.6374	0.6457	0.0046	0.0041	0.2876	0.6200	
753	6	9	0.0053	-0.0000	-0.0006	-0.0006	-0.0006	0.01	-1.01	0.0026	0.2876	0.6200	0.0053	0.0014	0.0006	0.1792	0.1770	0.6374	0.6457	0.0046	0.0041	0.2876	0.6200	

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																				
753	7	7	0.799	4.15	-0.000	0.0604	1.02	-0.85	-1.01	1.04	0.2739	0.6170	0.0087	0.0001	-0.0007	0.05047	0.0343	0.0018	0.0053	0.0042	0.2457	0.6179	
		8	0.0033	-0.0001	-0.0007	0.2515	1.1807	0.0429	0.0021	0.0053	0.0042	0.6179											
753	8	7	0.800	-0.04	-0.000	0.2434	1.02	-0.93	-1.01	0.97	0.2505	0.4509	0.0076	0.0003	-0.0017	0.7351	0.0010	0.0001	0.0006	0.0002	0.1347	0.4509	
		8	0.0055	-0.0001	-0.0017	0.0986	0.6703	0.0053	0.0001	0.0006	0.0006	0.4509											
754	1	7	0.800	-3.10	-0.000	0.2188	2.08	-0.11	-2.11	-0.07	0.2192	0.6434	0.0126	0.0005	-0.0014	0.6302	-0.0233	-0.0012	-0.0030	0.0030	0.2713	0.6434	
		8	0.0118	-0.0004	-0.0014	0.1992	0.6314	-0.0235	-0.0015	-0.0030	0.0030	0.6434											
754	2	7	0.799	-0.05	-0.000	0.2347	2.08	-0.06	-2.10	-0.03	0.2079	0.6417	0.0141	0.0006	-0.0014	0.6457	0.029	0.0002	0.0003	0.0003	0.4026	0.5436	
		8	0.0114	-0.0004	-0.0014	0.2079	0.6417	0.0002	-0.0001	-0.0003	0.0003	0.5436											
754	3	7	0.800	0.49	-0.000	0.2209	2.08	-0.04	-2.11	-0.01	0.2171	0.6529	0.0146	0.0006	-0.0014	0.6274	0.0068	0.0004	0.0005	0.0005	0.3428	0.6379	
		8	0.0146	0.0006	-0.0014	0.2171	0.6529	0.0042	0.0000	0.0005	0.0005	0.6379											
754	4	7	0.799	1.02	-0.000	0.2664	2.08	-0.03	-2.11	-0.01	0.2505	0.7167	0.0107	0.0005	-0.0015	0.6767	0.0115	0.0002	0.0010	0.0010	0.3121	0.6357	
		8	0.0140	0.0007	-0.0015	0.2505	0.7167	0.0080	0.0002	0.0010	0.0010	0.6357											
754	5	7	0.799	2.06	-0.000	0.2560	2.08	-0.01	-2.11	0.00	0.2348	0.6925	0.0108	0.0005	-0.0015	0.6576	0.0205	0.0007	0.0021	0.0021	0.2925	0.6327	
		8	0.0108	-0.0005	-0.0015	0.2348	0.6925	0.0168	0.0007	0.0021	0.0021	0.6327											
754	6	7	0.800	3.10	-0.000	0.2500	2.08	-0.00	-2.10	0.02	0.1893	0.6228	0.0139	0.0006	-0.0014	0.6440	0.0303	0.0016	0.0032	0.0032	0.2790	0.6288	
		8	0.0139	0.0006	-0.0014	0.1893	0.6228	0.0257	0.0012	0.0032	0.0032	0.6288											
754	7	7	0.800	4.13	-0.000	0.2009	2.08	0.02	-2.11	0.04	0.3184	0.9041	0.0086	0.0005	-0.0015	0.5995	0.0399	0.0017	0.0045	0.0045	0.2727	0.6215	
		8	0.0086	-0.0005	-0.0015	0.3184	0.9041	0.0362	0.0017	0.0045	0.0045	0.6215											
754	8	7	0.800	-0.03	-0.000	0.2656	2.08	-0.06	-2.10	-0.02	0.1843	0.5936	0.0137	0.0007	-0.0014	0.6845	0.0030	0.0002	0.0003	0.0003	0.3692	0.5163	
		8	0.0137	0.0007	-0.0014	0.1843	0.5936	0.0002	-0.0001	-0.0003	0.0003	0.5163											
755	1	7	0.800	-3.10	-0.000	0.2340	2.07	-1.91	-2.12	1.92	0.2437	0.7265	0.0114	0.0004	-0.0014	0.6563	0.0341	0.0018	0.0043	0.0043	0.2658	0.6370	
		8	0.0114	0.0004	-0.0014	0.2437	0.7265	0.0127	-0.0008	-0.0016	0.0016	0.6370											

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																			
755	2	7	0.800	-0.04	-0.00	0.2298	2.08	-1.85	-2.12	1.97	0.2609	0.7862	0.0132	0.0006	0.0016	0.2101	0.6389	-0.0052	-0.0002	-0.0008	0.2045	0.6691
		8	-0.0106	-0.0004	-0.0014																	
		9																				
755	3	7	0.800	0.45	-0.00	0.2087	2.08	-1.84	-2.11	1.97	0.2078	1.4754	0.0138	0.0005	0.0016	0.2083	0.6099	-0.0008	-0.0000	-0.0002	0.2276	0.6560
		8	-0.0104	-0.0004	-0.0013																	
		9																				
755	4	7	0.801	1.02	-0.00	0.2098	2.08	-1.83	-2.11	1.99	0.3938	0.4790	0.0139	0.0005	0.0016	0.2021	0.6078	0.0026	0.0002	0.0002	0.2346	0.6409
		8	-0.0105	-0.0004	-0.0013																	
		9																				
755	5	7	0.801	2.05	-0.00	0.1795	2.08	-1.80	-2.11	2.00	0.3201	0.6207	0.0142	0.0005	0.0016	0.1225	0.5708	0.0109	0.0007	0.0013	0.2450	0.6294
		8	-0.0121	-0.0002	-0.0011																	
		9																				
755	6	7	0.800	3.10	-0.00	0.2531	2.08	-1.78	-2.11	2.02	0.2887	0.6193	0.0130	0.0006	0.0016	0.1928	0.6446	0.0196	0.0011	0.0024	0.2460	0.6280
		8	-0.0099	-0.0003	-0.0012																	
		9																				
755	7	7	0.799	4.14	-0.00	0.2826	2.08	-1.76	-2.10	2.04	0.2822	0.6108	0.0121	0.0006	0.0011	0.1663	0.6006	0.0292	0.0016	0.0036	0.2470	0.6108
		8	-0.0099	-0.0003	-0.0011																	
		9																				
755	8	7	0.801	-0.01	-0.00	0.21857	2.07	-1.84	-2.11	1.96	0.2750	0.6368	0.0135	0.0005	0.0013	0.1857	0.6335	-0.0050	-0.0002	-0.0013	0.2069	0.6368
		8	-0.0106	-0.0003	-0.0013																	
		9																				
756	1	7	0.799	-3.10	-0.00	0.2550	5.07	-0.10	-5.07	-0.08	0.2702	0.6373	0.0288	0.0014	0.0036	0.2675	0.6504	-0.0220	-0.0011	-0.0032	0.2069	0.6373
		8	-0.0288	-0.0014	-0.0036																	
		9																				
756	2	7	0.798	-0.04	-0.00	0.2740	5.07	-0.05	-5.07	-0.03	0.4123	0.6406	0.0306	0.0016	0.0040	0.2768	0.6476	-0.0036	0.0002	-0.0001	0.2069	0.6406
		8	-0.0306	-0.0015	-0.0035																	
		9																				
756	3	7	0.800	0.49	-0.00	0.2526	5.07	-0.04	-5.07	-0.02	0.3178	0.6300	0.0315	0.0015	0.0040	0.2708	0.6388	-0.0082	0.0005	0.0010	0.2069	0.6300
		8	-0.0315	-0.0015	-0.0034																	
		9																				
756	4	7	0.798	1.00	-0.00	0.2805	5.08	-0.03	-5.07	-0.01	0.2953	0.6231	0.0278	0.0017	0.0040	0.2769	0.6493	-0.0128	0.0007	0.0015	0.2069	0.6231
		8	-0.0278	-0.0014	-0.0035																	
		9																				

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																
756	5	0.799	2.05	0.00	0.2558	5.07	-0.00	-5.07	-0.00	0.00	0.2841	0.6292	0.0027	0.0012	0.0006	0.0019	0.2837	0.6288
	6	0.312	0.0016	0.0040	0.2767	0.6411	0.0153	0.0012	0.0218	0.0012	0.0019	0.6292	0.0027	0.0012	0.0006	0.0019	0.2837	0.6288
	9	-0.0266	-0.0014	-0.0034	0.2767	0.6499	0.0153	0.0006	0.0153	0.0006	0.0019	0.6288	0.0019	0.0006	0.0012	0.0019	0.2837	0.6288
756	6	0.800	3.10	0.00	0.2444	5.07	0.00	-5.07	0.00	0.00	0.2760	0.6239	0.0039	0.0017	0.0030	0.0030	0.2777	0.6236
	8	0.316	0.0015	0.0039	0.3211	0.6299	0.0248	0.0012	0.0320	0.0012	0.0030	0.6239	0.0039	0.0017	0.0030	0.0030	0.2777	0.6236
	9	-0.0243	-0.0015	-0.0035	0.3211	0.7347	0.0248	0.0012	0.0248	0.0012	0.0030	0.6236	0.0030	0.0012	0.0030	0.0030	0.2777	0.6236
756	7	0.800	4.16	0.00	0.2501	5.07	0.02	-5.07	0.02	0.00	0.2694	0.6193	0.004	0.0022	0.0043	0.0043	0.2694	0.6193
	8	0.310	0.0015	0.0032	0.2623	0.6354	0.0353	0.0017	0.0413	0.0017	0.0043	0.6193	0.0043	0.0022	0.0043	0.0043	0.2694	0.6193
	9	-0.0260	-0.0013	-0.0032	0.2623	0.6251	0.0353	0.0017	0.0353	0.0017	0.0043	0.6193	0.0043	0.0017	0.0043	0.0043	0.2694	0.6193
756	8	0.799	-0.05	-0.00	0.2501	5.07	-0.05	-5.07	-0.05	0.00	0.3726	0.5944	-0.003	0.0002	-0.0001	0.0001	0.3726	0.5944
	8	0.313	0.0015	0.0039	0.2866	0.6309	0.0038	0.0002	0.0038	0.0002	0.0004	0.5944	0.0004	0.0002	0.0001	0.0001	0.3726	0.5944
	9	-0.0267	-0.0015	-0.0035	0.2866	0.6721	-0.0008	-0.0001	-0.0008	-0.0001	0.0001	0.5944	0.0001	-0.0001	-0.0001	0.0001	0.3726	0.5944
757	1	0.799	-3.11	-0.00	0.2805	5.06	-5.13	-5.06	-5.13	0.00	0.2707	0.5126	5.01	0.0026	0.0001	0.0001	0.2707	0.5126
	8	0.257	0.0014	0.0033	0.2598	0.6595	-0.0493	-0.0026	-0.0493	-0.0026	0.0060	0.5126	0.0060	0.0026	0.0001	0.0001	0.2707	0.5126
	9	-0.0263	-0.0013	-0.0033	0.2598	0.6260	0.0007	-0.0000	0.0007	-0.0000	0.0001	0.5126	0.0001	-0.0000	-0.0001	0.0001	0.2707	0.5126
757	2	0.800	-0.05	-0.00	0.3022	5.06	-5.07	-5.06	-5.07	0.00	0.2714	0.6524	5.06	0.0011	0.0033	0.0033	0.2714	0.6524
	8	0.279	0.0016	0.0037	0.3022	0.6645	-0.0213	-0.0012	-0.0213	-0.0012	0.0027	0.6524	0.0027	0.0011	0.0033	0.0033	0.2714	0.6524
	9	-0.0239	-0.0014	-0.0033	0.3022	0.6896	0.0254	0.0012	0.0254	0.0012	0.0033	0.6524	0.0033	0.0012	0.0033	0.0033	0.2714	0.6524
757	3	0.900	0.48	-0.00	0.2670	5.06	-5.06	-5.06	-5.06	0.00	0.2733	0.6760	5.07	0.0008	0.0039	0.0039	0.2733	0.6760
	8	0.290	0.0015	0.0037	0.2596	0.6418	-0.0302	-0.0015	-0.0302	-0.0015	0.0057	0.6760	0.0057	0.0008	0.0039	0.0039	0.2733	0.6760
	9	-0.0255	-0.0013	-0.0031	0.2596	0.6109	0.0302	0.0015	0.0302	0.0015	0.0039	0.6760	0.0039	0.0015	0.0039	0.0039	0.2733	0.6760
757	4	0.800	1.02	-0.00	0.2690	5.07	-5.05	-5.06	-5.05	0.00	0.2724	0.7084	5.08	0.0006	0.0045	0.0045	0.2724	0.7084
	8	0.290	0.0015	0.0037	0.2637	0.6424	-0.0112	-0.0006	-0.0112	-0.0006	0.0050	0.7084	0.0050	0.0006	0.0045	0.0045	0.2724	0.7084
	9	-0.0250	-0.0013	-0.0031	0.2637	0.6206	0.0354	0.0017	0.0354	0.0017	0.0045	0.7084	0.0045	0.0017	0.0045	0.0045	0.2724	0.7084
757	5	0.799	2.05	-0.00	0.2730	5.07	-5.03	-5.06	-5.03	0.00	0.3294	1.0116	5.10	0.0001	0.0055	0.0055	0.3294	1.0116
	8	0.288	0.0015	0.0037	0.2933	0.6485	-0.0024	-0.0001	-0.0024	-0.0001	0.0055	1.0116	0.0055	0.0001	0.0055	0.0055	0.3294	1.0116
	9	-0.0234	-0.0013	-0.0031	0.2933	0.6763	0.0044	0.0022	0.0044	0.0022	0.0055	1.0116	0.0055	0.0022	0.0055	0.0055	0.3294	1.0116
757	6	0.798	3.09	0.00	0.2537	5.06	-5.02	-5.06	-5.02	0.00	0.3170	0.5270	5.12	0.0002	0.0062	0.0062	0.3170	0.5270
	8	0.292	0.0014	0.0036	0.3080	0.6276	0.0044	0.0002	0.0044	0.0002	0.0062	0.5270	0.0062	0.0002	0.0062	0.0062	0.3170	0.5270
	9	-0.0224	-0.0013	-0.0032	0.3080	0.7151	0.0525	0.0026	0.0525	0.0026	0.0062	0.5270	0.0062	0.0026	0.0062	0.0062	0.3170	0.5270
757	7	0.800	4.14	-0.00	0.2388	5.06	-5.00	-5.05	-5.00	0.00	0.2823	0.5779	5.13	0.0007	0.0066	0.0066	0.2823	0.5779
	8	0.292	0.0013	0.0035	0.2725	0.6117	0.0125	0.0007	0.0125	0.0007	0.0066	0.5779	0.0066	0.0007	0.0066	0.0066	0.2823	0.5779
	9	-0.0228	-0.0012	-0.0029	0.2725	0.6527	0.0583	0.0030	0.0583	0.0030	0.0066	0.5779	0.0066	0.0030	0.0066	0.0066	0.2823	0.5779

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key														
757	8	0.800	-0.01	-0.00	5.06	-5.07	-5.06	5.07	-5.06	5.07	-5.06	5.07	-5.06	5.07	0.2747	0.6617
	8	0.0291	0.0014	0.0036	0.6200	-0.0211	-0.0011	-0.0033	-0.0011	-0.0033	-0.0011	-0.0033	-0.0011	-0.0033	0.2541	0.6485
	9	-0.0253	-0.0013	-0.0031	0.2448	0.6220	0.0255	0.0013	0.0033	0.0255	0.0013	0.0033	0.0255	0.0013	0.0033	
758	3	0.600	-3.12	-0.00	-0.02	-3.01	0.05	-3.01	0.05	-3.01	0.05	-3.01	0.05	-3.01	0.2684	0.6153
	8	0.0014	-0.0000	0.0002	0.7928	-0.0395	-0.0021	-0.0048	-0.0021	-0.0048	-0.0021	-0.0048	-0.0021	-0.0048	0.3160	0.6149
	9	-0.0015	0.0000	-0.0002	0.8220	-0.0374	-0.0023	-0.0046	-0.0023	-0.0046	-0.0023	-0.0046	-0.0023	-0.0046		
758	4	0.601	-1.07	-0.00	-0.03	-2.97	0.05	-2.97	0.05	-2.97	0.05	-2.97	0.05	-2.97	0.3504	0.6216
	8	0.0026	-0.0001	0.0002	0.4377	-0.0203	-0.0010	-0.0024	-0.0010	-0.0024	-0.0010	-0.0024	-0.0010	-0.0024	0.3575	0.6226
	9	-0.0029	0.0001	-0.0001	0.1815	-0.0195	-0.0013	-0.0024	-0.0013	-0.0024	-0.0013	-0.0024	-0.0013	-0.0024		
758	5	0.601	-0.06	-0.00	-0.03	-2.96	0.05	-2.96	0.05	-2.96	0.05	-2.96	0.05	-2.96	0.3300	0.6629
	8	0.0024	-0.0000	0.0003	0.8066	-0.0113	-0.0008	-0.0015	-0.0008	-0.0015	-0.0008	-0.0015	-0.0008	-0.0015	0.3802	0.6733
	9	-0.0007	0.0000	-0.0002	1.9350	-0.0113	-0.0008	-0.0015	-0.0008	-0.0015	-0.0008	-0.0015	-0.0008	-0.0015		
758	6	0.600	0.49	-0.00	-0.02	-2.94	0.05	-2.94	0.05	-2.94	0.05	-2.94	0.05	-2.94	0.2201	0.7442
	8	0.0026	-0.0000	0.0003	0.7351	-0.0070	-0.0003	-0.0010	-0.0003	-0.0010	-0.0003	-0.0010	-0.0003	-0.0010	0.4376	0.7188
	9	-0.0019	0.0001	-0.0002	0.5175	-0.0074	-0.0006	-0.0010	-0.0006	-0.0010	-0.0006	-0.0010	-0.0006	-0.0010		
758	7	0.600	0.97	-0.00	-0.02	-2.94	0.05	-2.94	0.05	-2.94	0.05	-2.94	0.05	-2.94	0.1456	0.9158
	8	0.0021	0.0000	0.0004	1.0796	-0.0033	-0.0004	-0.0006	-0.0004	-0.0006	-0.0004	-0.0006	-0.0004	-0.0006	0.5091	0.7299
	9	-0.0005	-0.0000	-0.0004	-4.2278	-0.0043	-0.0004	-0.0006	-0.0004	-0.0006	-0.0004	-0.0006	-0.0004	-0.0006		
758	8	0.600	3.09	-0.00	-0.03	-2.91	0.05	-2.91	0.05	-2.91	0.05	-2.91	0.05	-2.91	0.3501	0.6011
	8	0.0028	-0.0000	0.0003	0.5993	0.0113	0.0007	0.0012	0.0007	0.0012	0.0007	0.0012	0.0007	0.0012	0.1953	0.6106
	9	-0.0017	0.0001	-0.0001	0.4639	0.0101	0.0003	0.0003	0.0003	0.0003	0.0003	0.0003	0.0003	0.0003		
758	9	0.600	6.19	-0.00	-0.03	-2.85	0.05	-2.85	0.05	-2.85	0.05	-2.85	0.05	-2.85	0.3073	0.6121
	8	0.0022	-0.0001	0.0001	0.3714	0.0368	0.0022	0.0045	0.0022	0.0045	0.0022	0.0045	0.0022	0.0045	0.2643	0.6052
	9	-0.0018	0.0002	0.0000	-0.2623	0.0372	0.0019	0.0045	0.0019	0.0045	0.0019	0.0045	0.0019	0.0045		
758	10	0.600	9.27	-0.00	-0.03	-2.80	0.05	-2.80	0.05	-2.80	0.05	-2.80	0.05	-2.80	0.3000	0.5676
	8	0.0016	-0.0001	0.0001	0.5360	0.0601	0.0036	0.0069	0.0036	0.0069	0.0036	0.0069	0.0036	0.0069	0.2739	0.5655
	9	-0.0016	0.0001	-0.0000	0.1350	0.0614	0.0033	0.0069	0.0033	0.0069	0.0033	0.0069	0.0033	0.0069		
758	11	0.600	12.44	-0.00	-0.03	-2.78	0.05	-2.78	0.05	-2.78	0.05	-2.78	0.05	-2.78	0.2609	0.5365
	8	0.0011	-0.0001	0.0001	0.7611	0.0793	0.0041	0.0086	0.0041	0.0086	0.0041	0.0086	0.0041	0.0086	0.2433	0.5355
	9	-0.0020	0.0000	-0.0002	0.6856	0.0807	0.0039	0.0086	0.0039	0.0086	0.0039	0.0086	0.0039	0.0086		
758	12	0.601	-0.06	-0.00	-0.03	-2.95	0.05	-2.95	0.05	-2.95	0.05	-2.95	0.05	-2.95	0.3426	0.7005
	8	0.0025	-0.0000	0.0004	0.7745	-0.0108	-0.0008	-0.0015	-0.0008	-0.0015	-0.0008	-0.0015	-0.0008	-0.0015	0.3848	0.6923
	9	-0.0012	0.0000	-0.0002	1.0581	-0.0110	-0.0008	-0.0015	-0.0008	-0.0015	-0.0008	-0.0015	-0.0008	-0.0015		

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																													
759	1	7	0.600	-3.09	-0.00	0.6395	-0.02	-1.00	0.05	-1.05	0.2666	0.6219	8	0.0006	0.0000	0.0003	2.3646	-0.0279	-0.0014	-0.0034	0.3294	0.6229	9	-0.0009	0.0000	-0.0002	-0.1659	1.5869	-0.0264	-0.0017	-0.0032
759	2	7	0.600	-1.09	-0.00	0.4058	-0.02	-0.97	0.05	-1.01	0.2186	0.6691	8	0.0012	0.0000	0.0004	1.8547	-0.0098	-0.0004	-0.0013	0.3948	0.6497	9	-0.0005	0.0000	-0.0003	-0.3816	3.0336	-0.0093	-0.0007	-0.0012
759	3	7	0.600	-0.05	-0.00	-0.1328	-0.02	-0.96	0.05	-1.00	-0.0105	0.9346	8	0.0029	-0.0000	0.0003	0.6240	-0.0119	0.0000	-0.0003	0.5924	0.6528	9	-0.0016	0.0001	-0.0001	-0.3897	0.5309	-0.0028	-0.0003	-0.0003
759	4	7	0.600	0.49	-0.00	-0.0159	-0.02	-0.95	0.05	-1.00	0.7813	0.4687	8	0.0028	-0.0000	0.0003	0.7056	0.0015	0.0002	0.0001	0.9374	0.9567	9	-0.0026	0.0002	-0.0000	-0.4032	0.1064	-0.0005	-0.0001	0.0001
759	5	7	0.600	0.98	-0.00	0.1501	-0.02	-0.93	0.05	-0.99	0.4409	0.6267	8	0.0023	0.0000	0.0004	0.9733	0.0051	0.0004	0.0006	0.0904	0.6434	9	-0.0020	0.0001	-0.0001	-0.2875	0.4769	0.0039	0.0000	0.0005
759	6	7	0.600	3.11	-0.00	0.0221	-0.02	-0.90	0.05	-0.95	0.3252	0.6212	8	0.0025	0.0000	0.0003	0.7264	0.0217	0.0014	0.0027	0.2586	0.6269	9	-0.0014	0.0001	-0.0001	0.5719	0.3941	0.0202	0.0010	0.0025
759	7	7	0.601	6.18	0.00	-0.2173	-0.03	-0.84	0.05	-0.91	0.3058	0.5995	8	0.0014	-0.0000	0.0002	0.9456	0.0471	0.0028	0.0056	0.2675	0.6003	9	0.0003	0.0001	-0.0001	1.3276	-2.2973	0.0475	0.0025	0.0057
759	8	7	0.600	9.27	-0.00	0.0175	-0.03	-0.80	0.05	-0.87	0.2856	0.5590	8	0.0006	0.0000	0.0003	2.2770	0.0689	0.0039	0.0077	0.2675	0.5557	9	0.0001	0.0000	-0.0001	1.9033	-6.1089	0.0696	0.0037	0.0077
759	9	7	0.599	12.46	-0.00	-0.7117	-0.03	-0.79	0.05	-0.86	0.2282	0.5361	8	0.0008	-0.0001	0.0001	1.1720	0.0841	0.0038	0.0090	0.2232	0.5330	9	-0.0029	0.0001	-0.0001	-0.2156	0.2058	0.0855	0.0038	0.0091
759	10	7	0.599	-0.03	-0.00	-0.0610	-0.03	-0.96	0.05	-1.00	-0.0303	0.6753	8	0.0025	-0.0000	0.0003	0.7171	-0.0015	0.0000	-0.0003	0.0303	0.6753	9	-0.0008	0.0000	-0.0002	-0.4772	1.4340	-0.0023	-0.0003	-0.0003
760	1	7	0.600	-3.08	-0.00	-0.2714	-0.02	-0.10	0.04	-0.06	0.2629	0.6153	8	0.0012	-0.0000	0.0002	0.8787	-0.0230	0.0012	-0.0028	0.3412	0.6153	9	-0.0019	0.0000	-0.0002	-0.1387	0.6547	-0.0209	-0.0014	-0.0028

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																			
760	2	7	0.600	-1.07	-0.00	-0.03	-0.07	0.04	-0.03	-0.007	0.1350	0.7350	0.0027	-0.0001	0.0003	0.6049	-0.0051	-0.0001	-0.0007	0.3509	0.6957
		8	-0.0018	0.0000	-0.0001	0.4537	-0.0043	-0.0004	-0.0004	-0.0006											
		9																			
760	3	7	0.600	-0.03	-0.00	-0.02	-0.05	0.04	-0.02	-0.002	0.7034	0.4715	0.0030	-0.0001	0.0003	0.6160	0.0017	-0.0002	0.0001	0.3814	0.7978
		8	-0.0020	0.0000	-0.0000	0.2219	0.0010	-0.0000	-0.0000	-0.0001											
		9																			
760	4	7	0.600	0.48	-0.00	-0.02	-0.05	0.05	-0.02	-0.000	0.4593	0.6508	0.0017	0.0001	0.0004	1.3180	0.0052	-0.0001	0.0006	0.1619	0.7190
		8	-0.0016	0.0001	-0.0002	0.6846	0.0047	0.0001	0.0001	0.0006											
		9																			
760	5	7	0.599	0.99	-0.00	-0.02	-0.04	0.05	-0.02	-0.000	0.3838	0.6396	0.0020	0.0001	0.0004	1.1964	0.0094	-0.0003	0.0012	0.2009	0.6411
		8	-0.0016	0.0001	-0.0001	0.4720	0.0089	0.0003	0.0003	0.0011											
		9																			
760	6	7	0.600	3.11	-0.00	-0.03	-0.00	0.05	-0.03	-0.003	0.3142	0.6183	0.0028	-0.0001	0.0003	0.6330	0.0269	-0.0001	0.0033	0.2615	0.6206
		8	-0.0016	0.0001	-0.0001	0.3254	0.0257	0.0013	0.0013	0.0031											
		9																			
760	7	7	0.600	6.13	-0.00	-0.03	-0.04	0.05	-0.03	-0.000	0.3074	0.5921	0.0011	-0.0001	0.0002	1.2865	0.0513	-0.0002	0.0060	0.2731	0.5933
		8	-0.0005	0.0001	-0.0000	0.2151	0.0527	0.0028	0.0028	0.0062											
		9																			
760	8	7	0.599	9.29	-0.00	-0.02	-0.08	0.05	-0.02	-0.000	0.2807	0.5524	0.0014	-0.0002	0.0002	0.7443	0.0725	-0.0000	0.0080	0.2617	0.5452
		8	-0.0017	0.0002	-0.0000	0.0929	0.0729	0.0038	0.0038	0.0079											
		9																			
760	9	7	0.599	12.44	-0.00	-0.03	-0.06	0.05	-0.03	-0.000	0.2145	0.5377	0.0004	-0.0000	0.0002	2.9130	0.0854	-0.0002	0.0091	0.2110	0.5298
		8	-0.0016	0.0000	-0.0002	0.6884	0.0869	0.0036	0.0036	0.0092											
		9																			
760	10	7	0.599	-0.03	-0.00	-0.02	-0.05	0.04	-0.02	-0.001	0.5983	0.3948	0.0028	-0.0001	0.0003	0.6509	0.0021	-0.0000	0.0001	0.1963	0.6650
		8	-0.0018	0.0001	-0.0001	0.4537	0.0019	0.0000	0.0000	0.0001											
		9																			
761	1	7	0.600	-3.07	-0.00	-0.03	-0.96	0.05	-0.03	-0.000	0.2556	0.6266	0.0017	-0.0000	0.0002	0.6226	0.0168	-0.0010	0.0021	0.3565	0.6262
		8	-0.0013	0.0000	-0.0002	0.9313	-0.0153	-0.0010	-0.0010	-0.0019											
		9																			
761	2	7	0.599	-1.05	-0.00	-0.03	1.01	0.05	-0.03	-0.000	-5.4007	2.1628	0.0031	-0.0001	0.0002	0.4304	0.0001	-0.0001	0.0000	1.7476	0.6782
		8	-0.0014	0.0001	-0.0001	0.6699	-0.0005	-0.0001	-0.0001	-0.0000											
		9																			

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd																					
761	3	7	0.599	-0.04	-0.00	0.3941	-0.02	1.02	0.04	0.97	0.4106	0.6434										
		8	0.0017	0.0001	0.0004	0.9310	1.4252	0.0065	0.0005	0.0006	0.1699	0.7051										
		9	-0.0001	0.0000	-0.0003	-0.9310	12.8574	0.0056	0.0001	0.0007												
761	4	7	0.600	0.49	-0.00	0.2770	-0.02	1.02	0.04	0.97	0.3672	0.6457										
		8	0.0021	0.0001	0.0005	0.4448	1.2085	0.0109	0.0008	0.0114												
		9	-0.0001	0.0000	-0.0002	-0.4448	2.0357	0.0099	0.0004	0.0012												
761	5	7	0.599	0.99	-0.00	0.0137	-0.02	1.03	0.04	0.98	0.3429	0.6270										
		8	0.0027	0.0000	0.0004	0.4074	0.7806	0.0147	0.0010	0.0016												
		9	-0.0009	0.0000	-0.0002	-0.4074	1.2868	0.0132	0.0006	0.0017												
761	6	7	0.599	3.11	-0.00	0.2705	-0.03	1.07	0.05	1.01	0.3089	0.6218										
		8	0.0041	-0.0002	0.0002	0.4649	0.3068	0.0329	0.0020	0.0040												
		9	-0.0022	0.0000	-0.0000	-0.4649	0.0819	0.0314	0.0016	0.0038												
761	7	7	0.598	6.20	-0.00	0.4346	-0.03	1.12	0.05	1.06	0.3042	0.5795										
		8	0.0021	-0.0001	0.0001	0.8551	0.4335	0.0557	0.0033	0.0064												
		9	-0.0012	0.0002	0.0001	-0.8551	-0.3495	0.0569	0.0031	0.0065												
761	8	7	0.600	9.30	-0.00	0.3739	-0.03	1.15	0.05	1.09	0.2739	0.5449										
		8	0.0005	-0.0000	0.0002	0.7620	2.1726	0.0753	0.0041	0.0082												
		9	-0.0002	0.0000	-0.0002	-0.7620	-6.2395	0.0753	0.0038	0.0082												
761	9	7	0.600	12.43	-0.00	0.1683	-0.03	1.16	0.05	1.09	0.2006	0.5328										
		8	0.0002	-0.0000	0.0002	0.2081	1.4954	0.0884	0.0035	0.0094												
		9	-0.0009	0.0000	-0.0002	-0.2081	-1.4954	0.0884	0.0035	0.0094												
761	10	7	0.602	-0.03	-0.00	0.0739	-0.02	1.02	0.04	0.97	0.4330	0.6658										
		8	0.0021	0.0000	0.0004	0.3146	0.9904	0.0064	0.0005	0.0007												
		9	-0.0016	0.0001	-0.0002	-0.3146	0.6310	0.0058	0.0001	0.0007												
762	1	7	0.601	-3.07	-0.00	0.2334	-0.02	0.96	0.04	3.03	0.1841	0.6827										
		8	0.0010	0.0000	0.0003	0.2491	0.6635	-0.0051	-0.0004	-0.0006												
		9	-0.0016	0.0000	-0.0002	-0.2491	-0.6635	-0.0051	-0.0004	-0.0006												
762	2	7	0.601	-1.08	-0.00	0.1314	-0.02	0.99	0.05	3.06	0.4022	0.6801										
		8	0.0017	0.0000	0.0004	0.3793	1.0017	0.0079	0.0003	0.0010												
		9	-0.0016	0.0001	-0.0001	-0.3793	0.5989	0.0086	0.0003	0.0011												
762	3	7	0.602	-0.02	-0.00	0.6734	-0.02	3.01	0.05	3.08	0.3419	0.6384										
		8	0.0011	0.0001	0.0005	0.3666	2.0748	0.0163	0.0011	0.0020												
		9	-0.0006	0.0000	-0.0002	-0.3666	-2.0748	0.0163	0.0007	0.0020												

See Key

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																			
762	7	9	0.601	-0.0000	0.51	-0.0003	-0.0070	-0.3875	-0.003	0.9425	0.0210	3.02	0.04	0.0026	3.09	0.3262	0.6337					
			-0.0013	-0.0001	-0.0001	0.0003	-0.3875	0.8073	0.0207	0.0010	0.0026	0.2593	0.6421									
762	5	9	0.601	-0.0000	0.99	-0.0004	-0.1189	-0.4193	-0.003	0.5815	3.03	0.05	3.09	0.3215	0.6389							
			-0.0018	-0.0001	-0.0001	0.0004	-0.4193	0.3484	0.0250	0.0012	0.0031	0.2571	0.6274									
762	6	9	0.601	-0.0000	3.14	-0.0003	-0.0679	-0.7623	-0.003	0.6136	3.07	0.05	3.13	0.3065	0.6155							
			-0.0008	-0.0001	-0.0001	0.0003	-0.7623	1.0078	0.0436	0.0022	0.0052	0.2638	0.6114									
762	7	9	0.600	-0.0028	6.20	-0.0001	-0.4826	-0.9785	-0.003	0.2453	3.11	0.05	3.17	0.2871	0.5698							
			-0.0011	-0.0002	0.0000	0.0001	-0.9785	-0.1557	0.0664	0.0035	0.0074	0.2656	0.5620									
762	8	9	0.601	-0.0003	9.30	-0.0002	-0.3341	-0.8498	-0.003	1.0020	3.13	0.05	3.19	0.2505	0.5402							
			-0.0003	-0.0000	-0.0000	0.0002	-0.8498	-2.9427	0.0827	0.0038	0.0088	0.2342	0.5362									
762	9	9	0.601	-0.0018	12.43	-0.0001	-0.4788	-0.3569	-0.001	0.4814	3.12	0.05	3.18	0.1813	0.5371							
			-0.0026	-0.0001	-0.0000	0.0000	-0.3569	0.0622	0.0870	0.0031	0.0091	0.1813	0.5253									
762	10	9	0.601	-0.0015	0.0000	-0.0004	0.3451	-0.4692	-0.002	1.5301	3.01	0.04	3.08	0.3358	0.6318							
			-0.0003	0.0000	-0.0000	0.0003	-0.4692	4.3516	0.0168	0.0007	0.0020	0.2389	0.6310									
763	1	9	0.601	-0.0010	3.05	-0.0003	0.0046	-0.2738	-0.003	1.4785	6.00	0.04	6.01	0.4258	0.6931							
			-0.0018	0.0001	-0.0001	0.0003	-0.2738	0.5130	0.0082	0.0003	0.0011	0.2057	0.7137									
763	2	9	0.601	-0.0018	1.05	-0.0004	0.0694	-0.4513	-0.003	1.1413	6.04	0.05	6.04	0.3243	0.6536							
			-0.0011	0.0000	-0.0001	0.0004	-0.4513	0.8964	0.0243	0.0012	0.0031	0.2571	0.6421									
763	3	9	0.601	-0.0034	0.01	-0.0003	-0.2069	-0.3263	-0.003	0.4736	6.06	0.04	6.06	0.3081	0.6400							
			-0.0022	-0.0001	-0.0001	0.0003	-0.3263	0.2924	0.0330	0.0017	0.0042	0.2617	0.6379									
763	4	9	0.601	-0.0032	0.50	-0.0003	-0.1324	-0.3709	-0.003	0.5551	6.07	0.04	6.07	0.3054	0.6311							
			-0.0026	-0.0001	-0.0000	0.0003	-0.3709	0.0964	0.0375	0.0019	0.0047	0.2608	0.6270									

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																			
763	5	7	0.601	1.02	-0.0003	-0.0607	-0.003	6.08	0.04	6.08	0.3060	0.6245	0.0029	0.0000	0.0003	0.6802	0.0425	0.0026	0.0053	0.2658	0.6251
763	6	9	-0.0012	-0.0000	-0.0002	-0.1939	1.0844	0.0418	0.0022	0.0052											
763	6	9	0.600	3.15	-0.0003	-0.1323	0.5399	6.11	0.04	6.11	0.2982	0.5852	0.0030	0.0000	0.0003	0.5399	0.0573	0.0034	0.0067	0.2720	0.5819
763	6	9	-0.0003	0.0000	-0.0002	-1.4126	3.6635	0.0573	0.0031	0.0066											
763	7	7	0.601	6.22	-0.0003	-0.0620	-0.003	6.15	0.05	6.15	0.2700	0.5510	0.0011	0.0000	0.0003	1.3921	0.0760	0.0041	0.0063	0.2524	0.5440
763	6	9	-0.0004	0.0001	-0.0000	-1.9748	0.6275	0.0761	0.0038	0.0082											
763	8	7	0.600	9.31	-0.0001	-0.5527	-0.003	6.14	0.04	6.14	0.1992	0.5428	0.0014	0.0001	0.0001	0.6661	0.0868	0.0034	0.0094	0.1958	0.5315
763	6	9	-0.0014	0.0001	-0.0000	-0.4624	0.2155	0.0879	0.0034	0.0093											
763	9	7	0.600	12.44	-0.0002	-0.5310	-0.003	6.13	0.05	6.13	0.1623	0.5400	0.0010	0.0001	0.0002	1.0766	0.0854	0.0027	0.0093	0.1495	0.5328
763	6	9	-0.0031	0.0001	-0.0000	-0.2926	0.0704	0.0876	0.0026	0.0093											
763	10	7	0.602	0.00	-0.0004	0.1744	-0.002	6.06	0.05	6.05	0.3083	0.6371	0.0020	0.0001	0.0001	0.9544	0.0333	0.0020	0.0042	0.3027	0.6555
764	1	8	0.602	0.00	-0.0001	-0.3570	0.5378	0.0328	0.0017	0.0041											
764	8	9	-0.0016	0.0000	-0.0001	0.1744	0.5378	0.0328	0.0017	0.0041											
764	1	7	0.602	3.01	-0.0002	-0.3206	-0.003	9.09	0.04	9.09	0.3207	0.6555	0.0023	0.0001	0.0002	0.5091	0.0228	0.0014	0.0029	0.2525	0.6555
764	8	9	-0.0024	0.0001	-0.0000	-0.3358	0.2017	0.0252	0.0011	0.0050											
764	2	7	0.602	0.98	-0.0004	0.1015	-0.003	9.13	0.05	9.13	0.3016	0.6328	0.0017	0.0002	0.0002	1.2211	0.0411	0.0024	0.0051	0.3016	0.6328
764	7	6	0.601	0.00	-0.0004	-0.3311	1.4545	0.0410	0.0021	0.0051											
764	6	9	-0.0009	0.0000	-0.0002	-0.1147	1.4545	0.0410	0.0021	0.0051											
764	3	7	0.601	0.00	-0.0003	-0.3137	-0.003	9.14	0.04	9.14	0.3061	0.6091	0.0029	0.0000	0.0003	0.6439	0.0486	0.0029	0.0059	0.2680	0.6146
764	8	9	-0.0011	0.0000	-0.0002	-0.1147	1.1074	0.0486	0.0026	0.0059											
764	4	7	0.601	0.52	-0.0004	0.1050	-0.002	9.15	0.04	9.15	0.3024	0.5974	0.0022	0.0000	0.0004	0.9963	0.0525	0.0031	0.0063	0.2705	0.6006
764	6	9	-0.0020	0.0001	-0.0001	-0.2809	0.4343	0.0526	0.0028	0.0063											
764	5	7	0.601	1.05	-0.0005	0.1026	-0.002	9.16	0.04	9.16	0.2959	0.5873	0.0029	0.0001	0.0001	0.8799	0.0560	0.0033	0.0065	0.2704	0.5917
764	8	9	-0.0012	0.0001	-0.0001	-0.4660	0.7780	0.0554	0.0029	0.0065											

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																												
764	6	7	0.601	3.17	-0.000	0.6339	-0.032	9.19	0.05	9.10	0.2780	0.5673	8	0.0013	0.0001	0.0004	1.7335	0.0708	0.0039	0.0080	0.2611	0.5615	9	-0.0017	0.0001	-0.0001	0.4420	0.0704	0.0036	0.0079
764	7	7	0.602	6.23	-0.000	3.3879	-0.032	9.19	0.06	9.10	0.2348	0.5441	8	0.0001	0.0001	0.0004	10.3436	0.0824	0.0038	0.0089	0.2243	0.5356	9	-0.0005	0.0001	-0.0000	0.4266	0.0836	0.0037	0.0089
764	8	7	0.602	9.30	-0.000	-0.5244	-0.03	9.18	0.05	9.10	0.1763	0.5424	8	0.0021	0.0002	0.0001	-0.3699	0.0858	0.0030	0.0093	0.1777	0.5281	9	-0.0038	0.0003	0.0002	-0.3221	0.0878	0.0031	0.0092
764	9	7	0.601	12.45	-0.000	-0.8062	-0.03	9.17	0.04	9.09	0.1440	0.5502	8	0.0004	0.0000	0.0002	2.7127	0.0867	0.0024	0.0095	0.1337	0.5597	9	-0.0021	0.0001	-0.0001	0.4083	0.0894	0.0023	0.0096
764	10	7	0.601	0.01	-0.000	-0.1058	-0.03	9.14	0.04	9.06	0.3063	0.6077	8	0.0030	0.0000	0.0003	0.6438	0.0487	0.0029	0.0059	0.3063	0.6130	9	-0.0024	0.0001	-0.0000	0.1421	0.0484	0.0025	0.0059
765	1	7	0.602	3.01	-0.000	-0.1300	-0.03	12.05	0.05	12.08	0.3018	0.6390	8	0.0016	0.0000	0.0003	0.9518	0.0393	0.0023	0.0050	0.2615	0.6447	9	-0.0026	0.0001	-0.0000	0.1637	0.0406	0.0021	0.0052
765	2	7	0.601	1.02	-0.000	-0.0029	-0.03	12.09	0.05	12.10	0.2977	0.5973	8	0.0016	0.0000	0.0003	1.1449	0.0543	0.0032	0.0064	0.2655	0.6000	9	-0.0007	0.0000	-0.0003	2.3060	0.0553	0.0029	0.0066
765	3	7	0.601	0.02	-0.000	-0.1048	-0.03	12.10	0.04	12.12	0.2830	0.5851	8	0.0027	0.0000	0.0003	0.6805	0.0621	0.0035	0.0072	0.2619	0.5847	9	-0.0020	0.0001	-0.0001	0.3439	0.0623	0.0032	0.0072
765	4	7	0.600	0.55	-0.000	-0.1834	-0.03	12.11	0.05	12.11	0.2804	0.5806	8	0.0036	0.0001	0.0003	0.4851	0.0660	0.0037	0.0076	0.2604	0.5778	9	-0.0031	0.0002	-0.0000	0.0536	0.0661	0.0034	0.0076
765	5	7	0.601	1.05	-0.000	-0.2433	-0.02	12.11	0.05	12.12	0.2766	0.5709	8	0.0021	0.0001	0.0004	1.1420	0.0695	0.0038	0.0079	0.2599	0.5717	9	-0.0013	0.0000	-0.0002	0.8561	0.0693	0.0036	0.0079
765	6	7	0.600	3.17	-0.000	-0.0114	-0.02	12.13	0.05	12.15	0.2557	0.5472	8	0.0025	0.0000	0.0004	0.8222	0.0797	0.0040	0.0086	0.2403	0.5417	9	-0.0008	0.0001	-0.0001	0.9347	0.0799	0.0038	0.0086

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key															
765	7	7	0.600	6.24	-0.00	-0.3233	-0.03	12.11	0.05	12.14	0.1872	0.5485						
	8	8	0.0014	-0.0000	0.0002	5.3160	0.8454	0.0881	0.0033	0.0096	0.1881	0.5380						
	9	9	0.0001	0.0001	-0.0000	-0.0000	-2.8947	0.0899	0.0033	0.0096	0.1881	0.5380						
765	8	7	0.600	9.31	-0.00	-0.6742	-0.03	12.11	0.05	12.12	0.1607	0.5432						
	8	8	0.0011	-0.0001	0.0001	-0.4328	0.8202	0.0857	0.0027	0.0093	0.1515	0.5318						
	9	9	-0.0023	0.0002	0.0000	-0.0000	-0.0733	0.0884	0.0026	0.0094	0.1515	0.5318						
765	9	7	0.600	12.46	-0.00	-1.1988	-0.03	12.11	0.04	12.12	0.1384	0.5527						
	8	8	0.0003	-0.0000	0.0002	0.9300	3.8805	0.0898	0.0024	0.0099	0.1259	0.5414						
	9	9	-0.0014	0.0000	-0.0002	-0.0000	0.9226	0.0923	0.0023	0.0100	0.1259	0.5414						
765	10	7	0.601	0.00	-0.00	0.3706	-0.02	12.09	0.05	12.12	0.2823	0.5838						
	8	8	0.0017	0.0001	0.0005	-0.5511	4.643	0.0620	0.0035	0.0072	0.2823	0.5838						
	9	9	-0.0087	0.0008	-0.0002	-0.0000	1.5619	0.0625	0.0032	0.0072	0.2823	0.5838						
766	1	7	0.600	-3.00	-0.00	1.9757	-0.03	15.11	0.05	15.10	0.2970	0.6048						
	8	8	0.0002	0.0001	0.0003	-0.3387	6.5617	0.0531	0.0031	0.0064	0.2614	0.6035						
	9	9	-0.0018	0.0001	-0.0001	0.0000	0.4119	0.0552	0.0028	0.0066	0.2614	0.6035						
766	2	7	0.601	-0.96	-0.00	0.3088	-0.03	15.14	0.04	15.13	0.2751	0.5780						
	8	8	0.0012	0.0000	0.0004	-0.5725	7.673	0.0687	0.0037	0.0079	0.2547	0.5744						
	9	9	-0.0004	0.0000	-0.0002	0.0000	2.8272	0.0700	0.0035	0.0080	0.2547	0.5744						
766	3	7	0.600	0.06	-0.00	0.0040	-0.03	15.15	0.05	15.13	0.2598	0.5673						
	8	8	0.0021	0.0000	0.0004	-0.2951	0.234	0.0749	0.0040	0.0084	0.2537	0.5617						
	9	9	-0.0028	0.0001	-0.0001	0.0000	0.4109	0.0753	0.0038	0.0084	0.2537	0.5617						
766	4	7	0.601	0.55	-0.00	0.4704	-0.02	15.15	0.04	15.14	0.2551	0.5597						
	8	8	0.0016	0.0001	0.0005	-0.2703	5.796	0.0774	0.0041	0.0085	0.2505	0.5538						
	9	9	-0.0011	0.0000	-0.0002	0.0000	1.1419	0.0772	0.0038	0.0085	0.2505	0.5538						
766	5	7	0.600	1.05	-0.00	0.0470	-0.03	15.16	0.05	15.13	0.2529	0.5529						
	8	8	0.0027	-0.0000	0.0003	-0.3510	7.139	0.0793	0.0041	0.0087	0.2423	0.5459						
	9	9	-0.0022	0.0001	-0.0001	0.0000	0.2884	0.0797	0.0038	0.0087	0.2423	0.5459						
766	6	7	0.600	3.18	-0.00	0.0513	-0.02	15.15	0.05	15.14	0.2119	0.5467						
	8	8	0.0022	0.0000	0.0003	1.1874	8.199	0.0846	0.0035	0.0092	0.2079	0.5372						
	9	9	0.0002	0.0000	-0.0002	0.0000	-5.2181	0.0858	0.0035	0.0092	0.2079	0.5372						
766	7	7	0.600	6.24	-0.00	-2.4263	-0.03	15.14	0.05	15.12	0.1688	0.5494						
	8	8	0.0017	-0.0001	0.0002	-0.4164	5.702	0.0864	0.0029	0.0093	0.1688	0.5494						
	9	9	0.0002	0.0001	-0.0001	0.0000	-2.7508	0.0883	0.0030	0.0093	0.1688	0.5494						

TABLE VI

CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key															
766	8	7	0.600	9.31	-0.00	0.4655	10.8985	-0.03	15.14	0.04	15.12	0.1435	0.5493				
		8	0.0001	0.0000	0.0003	0.2104	-1.1185	0.0911	0.0866	0.0025	0.0097	0.1319	0.5390				
		9	0.0019	-0.0000	-0.0004	-0.2104	-1.1185	0.0911	0.0866	0.0024	0.0098	0.1319	0.5390				
766	9	7	0.601	12.46	-0.00	-1.7756	-12.1330	0.0935	15.14	0.04	15.12	0.1331	0.5536				
		8	-0.0001	0.0000	0.0003	-0.1267	2.2183	0.0956	0.0935	0.0024	0.0103	0.1214	0.5448				
		9	-0.0006	0.0000	-0.0002	-0.1267	2.2183	0.0956	0.0935	0.0023	0.0104	0.1214	0.5448				
766	10	7	0.600	0.02	-0.00	0.3970	1.5620	0.0753	15.15	0.04	15.13	0.2686	0.5648				
		8	0.0016	0.0001	0.0005	0.3395	0.3439	0.0753	0.0750	0.0040	0.0084	0.2517	0.5580				
		9	-0.0020	0.0001	-0.0001	-0.3395	0.3439	0.0753	0.0750	0.0037	0.0084	0.2517	0.5580				
767	1	7	0.602	-3.07	-0.00	0.2952	-2.99	2.95	3.00	2.94	-3.03	0.1689	0.6423				
		8	-0.0118	-0.0007	-0.0011	0.2479	0.5011	-0.0350	0.0149	-0.0002	-0.0010	0.3197	0.6247				
		9	0.0131	0.0006	0.0009	0.2479	0.5011	-0.0350	0.0149	-0.0022	-0.0043	0.3197	0.6247				
767	2	7	0.601	-0.05	-0.00	0.3620	-2.99	3.00	3.00	2.94	-2.98	0.3463	0.6313				
		8	-0.0101	-0.0007	-0.0011	0.4104	0.6299	-0.0103	0.0149	0.0010	0.0018	0.3463	0.6313				
		9	0.0103	0.0008	0.0013	0.4104	0.6299	-0.0103	0.0149	-0.0008	-0.0013	0.3463	0.6313				
767	3	7	0.601	0.45	-0.00	0.3839	-2.99	3.01	3.01	2.94	-2.98	0.3360	0.6399				
		8	-0.0093	-0.0007	-0.0010	0.3472	0.5792	-0.0063	0.0194	0.0013	0.0024	0.4686	0.7482				
		9	0.0113	0.0007	0.0012	0.3472	0.5792	-0.0063	0.0194	-0.0005	-0.0009	0.4686	0.7482				
767	4	7	0.602	0.99	-0.00	0.4880	-2.99	3.02	3.02	2.94	-2.97	0.3299	0.6394				
		8	-0.0084	-0.0008	-0.0011	0.4798	0.6671	-0.0035	0.0235	0.0015	0.0030	0.5629	0.7577				
		9	0.0094	0.0009	0.0013	0.4798	0.6671	-0.0035	0.0235	-0.0003	-0.0005	0.5629	0.7577				
767	5	7	0.602	2.04	-0.00	0.3334	-2.99	3.04	3.04	2.94	-2.96	0.3090	0.6235				
		8	-0.0098	-0.0006	-0.0010	0.4565	0.5302	0.0328	0.0328	0.0020	0.0040	0.0250	0.5334				
		9	0.0098	0.0009	0.0013	0.4565	0.5302	0.0328	0.0328	0.0000	0.0003	0.0250	0.5334				
767	6	7	0.600	3.08	-0.00	0.4620	-2.99	3.06	3.06	2.94	-2.94	0.3037	0.6118				
		8	-0.0091	-0.0008	-0.0011	0.5215	0.6392	0.0417	0.0417	0.0025	0.0051	0.2073	0.6021				
		9	0.0091	0.0009	0.0014	0.5215	0.6392	0.0417	0.0417	0.0004	0.0013	0.2073	0.6021				
767	7	7	0.600	4.11	-0.00	0.4448	-3.00	3.08	3.08	2.94	-2.93	0.3060	0.5931				
		8	-0.0095	-0.0008	-0.0012	0.3546	0.6341	0.0495	0.0495	0.0030	0.0058	0.2484	0.5997				
		9	0.0118	0.0008	0.0012	0.3546	0.6341	0.0495	0.0495	0.0009	0.0023	0.2484	0.5997				
767	8	7	0.600	-0.07	-0.00	0.3165	-2.99	3.00	3.00	2.94	-2.98	0.3329	0.6167				
		8	-0.0104	-0.0006	-0.0010	0.3915	0.5015	-0.0154	0.0154	-0.0010	-0.0019	0.4052	0.6984				
		9	0.0107	0.0008	0.0012	0.3915	0.5015	-0.0154	0.0154	-0.0008	-0.0013	0.4052	0.6984				

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key													
768	1	7	0.601	-3.09	-0.00	0.4463	-3.00	-0.11	2.94	-0.06	0.2673	0.6295			
		8	-0.0114	-0.0010	-0.0014	0.4463	0.6540	-0.0238	-0.0012	-0.0030	0.3454	0.6339			
		9	0.0107	0.0009	0.0014	0.4299	0.6788	-0.0196	-0.0013	-0.0024					
768	2	7	0.600	-0.04	-0.00	0.4802	-3.00	-0.06	2.94	-0.02	0.9371	0.0578			
		8	-0.0099	-0.0009	-0.0013	0.4802	0.6790	0.0010	0.0001	0.0000	-0.1645	0.6137			
		9	0.0101	0.0009	0.0014	0.4682	0.7401	0.0019	-0.0000	0.0002					
768	3	7	0.600	0.46	-0.00	0.3394	-2.99	-0.04	2.94	-0.01	0.4195	0.5375			
		8	-0.0105	-0.0007	-0.0011	0.3394	0.5512	0.0049	0.0004	0.0005	0.1641	0.6045			
		9	0.0121	0.0008	0.0013	0.3536	0.5522	0.0057	0.0001	0.0007					
768	4	7	0.600	1.00	-0.00	0.4000	-2.99	-0.04	2.94	-0.00	0.3915	0.6119			
		8	-0.0101	-0.0008	-0.0012	0.4000	0.6071	0.0083	0.0006	0.0010	0.1970	0.6170			
		9	0.0122	0.0008	0.0013	0.3568	0.5506	0.0096	0.0003	0.0011					
768	5	7	0.600	2.04	-0.00	0.3108	-2.99	-0.02	2.94	0.00	0.3370	0.6176			
		8	-0.0109	-0.0006	-0.0011	0.3108	0.5327	0.0167	0.0011	0.0020	0.2371	0.6083			
		9	0.0109	0.0009	0.0015	0.4489	0.7002	0.0177	0.0008	0.0021					
768	6	7	0.601	3.08	-0.00	0.3839	-2.99	-0.01	2.94	0.01	0.3234	0.6190			
		8	-0.0102	-0.0007	-0.0012	0.3839	0.5927	0.0251	0.0016	0.0031	0.2652	0.6239			
		9	0.0096	0.0011	0.0017	0.5753	0.9106	0.0261	0.0013	0.0032					
768	7	7	0.600	4.12	-0.00	0.2533	-2.99	0.00	2.94	0.04	0.3097	0.6116			
		8	-0.0123	-0.0006	-0.0011	0.2533	0.4824	0.0343	0.0021	0.0042	0.2636	0.6107			
		9	0.0113	0.0010	0.0016	0.4552	0.7093	0.0358	0.0018	0.0043					
768	8	7	0.603	-0.03	-0.00	0.4266	-3.00	-0.05	2.94	-0.02	0.8842	0.5223			
		8	-0.0101	-0.0008	-0.0012	0.4266	0.6373	0.0012	0.0002	0.0000	-0.0800	0.2057			
		9	0.0112	0.0008	0.0014	0.3949	0.6267	0.0022	-0.0000	0.0002					
769	1	7	0.601	-3.09	-0.00	0.0268	0.48	-0.10	0.48	-0.06	0.2627	0.6198			
		8	-0.0037	0.0000	0.0005	0.0268	0.6716	-0.0229	-0.0012	-0.0028	0.3393	0.6294			
		9	-0.0057	0.0000	-0.0003	-0.0179	0.2854	-0.0210	-0.0014	-0.0026					
769	2	7	0.601	-0.06	-0.00	0.0662	0.49	-0.05	0.48	-0.02	0.7035	0.4156			
		8	-0.0051	0.0000	0.0006	0.0662	0.6383	0.0018	0.0002	0.0001	-0.5018	0.9233			
		9	-0.0041	-0.0000	-0.0004	0.0589	0.5191	0.0008	-0.0000	0.0001					
769	3	7	0.600	0.45	-0.00	0.0364	0.49	-0.05	0.49	-0.01	0.4235	0.6062			
		8	-0.0053	0.0000	0.0006	0.0364	0.5820	0.0057	0.0004	0.0006	0.1482	0.6804			
		9	-0.0045	-0.0000	-0.0004	0.0611	0.5111	0.0047	0.0001	0.0006					

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																												
769	4	7	0.601	1.00	-0.00	0.2143	0.49	-0.04	-0.49	-0.01	0.3791	0.6232	8	0.0044	0.0001	0.0007	0.1115	0.8141	0.0094	0.0007	0.0011	0.1670	0.6135	9	-0.0035	-0.0000	-0.0005	0.0006	0.0003	0.0010
769	5	7	0.601	2.05	-0.00	0.2216	0.49	-0.02	-0.48	0.00	0.3431	0.6251	8	0.0048	0.0002	0.0007	0.1112	0.7620	0.0179	0.0012	0.0022	0.2411	0.6108	9	-0.0035	-0.0000	-0.0005	0.0166	0.0008	0.0020
769	6	7	0.601	3.10	-0.00	0.0954	0.49	-0.00	-0.48	0.02	0.3202	0.6190	8	0.0050	0.0000	0.0006	0.0138	0.6383	0.0265	0.0017	0.0032	0.2627	0.6108	9	-0.0037	-0.0000	-0.0003	0.0254	0.0013	0.0031
769	7	7	0.600	4.11	-0.00	0.3249	0.49	0.01	-0.49	0.03	0.3101	0.6181	8	0.0034	0.0002	0.0006	0.0282	0.9442	0.0353	0.0021	0.0043	0.2863	0.6153	9	-0.0042	0.0000	-0.0003	0.0344	0.0018	0.0042
769	8	7	0.600	-0.01	-0.00	0.2830	0.49	-0.05	-0.48	-0.02	0.7367	0.4219	8	0.0038	0.0002	0.0007	0.2082	0.9195	0.0018	0.0002	0.0001	0.4016	0.5210	9	-0.0030	-0.0001	-0.0005	0.0011	-0.0000	0.0001
770	1	7	0.600	-3.08	-0.00	0.1980	0.49	-0.55	-0.48	0.41	0.2645	0.6285	8	0.0028	0.0001	0.0005	0.1351	0.9760	-0.0251	-0.0013	-0.0023	0.3463	0.6298	9	-0.0037	-0.0001	-0.0004	0.0183	-0.0012	0.0023
770	2	7	0.600	-0.05	-0.00	0.0743	0.49	-0.50	-0.48	0.45	15.0517	-7.8946	8	0.0049	0.0000	0.0006	0.1972	0.6493	0.0000	0.0001	0.0003	0.0703	0.6539	9	-0.0031	-0.0001	-0.0005	0.0029	0.0000	0.0003
770	3	7	0.599	0.44	-0.00	0.2298	0.49	-0.50	-0.48	0.46	0.5505	0.5804	8	0.0045	0.0002	0.0007	0.1178	0.7981	0.0033	0.0003	0.0003	0.1899	0.6710	9	-0.0033	-0.0000	-0.0004	0.0066	0.0002	0.0008
770	4	7	0.600	1.03	-0.00	0.2343	0.49	-0.49	-0.48	0.46	0.4111	0.6260	8	0.0045	0.0002	0.0007	0.1186	0.8072	0.0074	0.0004	0.0013	0.2134	0.6241	9	-0.0035	-0.0000	-0.0004	0.0107	0.0004	0.0013
770	5	7	0.600	2.05	-0.00	0.1202	0.49	-0.47	-0.48	0.49	0.3406	0.6113	8	0.0053	0.0001	0.0006	0.1690	0.6459	0.0156	0.0010	0.0019	0.2558	0.6244	9	-0.0026	-0.0000	-0.0005	0.0184	0.0009	0.0023
770	6	7	0.600	3.10	-0.00	0.1622	0.49	-0.45	-0.48	0.50	0.3252	0.6194	8	0.0045	0.0001	0.0006	0.0632	0.7124	0.0245	0.0015	0.0030	0.2603	0.6092	9	-0.0035	-0.0000	-0.0004	0.0282	0.0014	0.0034

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																				
772	1	7	0.600	-3.08	-0.00	0.2313	1.01	-1.00	-1.02	0.92	0.2635	0.6234	0.0053	0.0002	-0.0007	0.1334	0.7802	-0.0275	-0.0014	-0.0034	0.3530	0.6278	
		8	-0.0062	-0.0001	0.0008	0.0000	0.5709	-0.0161	-0.0011	-0.0020													
		9																					
772	2	7	0.602	-0.08	-0.00	0.3185	1.02	-0.95	-1.02	0.97	-0.1246	0.9318	0.0060	0.0003	-0.0007	0.1432	0.8641	-0.0016	0.0000	-0.0003	0.1680	0.6707	
		8	-0.0061	-0.0001	0.0010	0.0000	0.5806	-0.0052	0.0001	0.0007													
		9																					
772	3	7	0.601	0.46	-0.00	0.1284	1.01	-0.93	-1.02	0.98	0.7240	0.4086	0.0073	0.0001	-0.0007	0.1530	0.6161	0.0018	0.0002	0.0012	0.1999	0.6328	
		8	-0.0061	-0.0001	0.0009	0.0000	0.5826	-0.0097	0.0003	0.0012													
		9																					
772	4	7	0.601	0.99	-0.00	0.1049	1.01	-0.93	-1.01	0.98	0.4713	0.6334	0.0079	0.0001	-0.0006	0.1389	0.5797	0.0052	0.0004	0.0016	0.2278	0.6306	
		8	-0.0062	-0.0001	0.0009	0.0000	0.5528	-0.0130	0.0005	0.0016													
		9																					
772	5	7	0.600	2.03	-0.00	0.3026	1.02	-0.91	-1.02	0.99	0.3602	0.6217	0.0063	0.0003	-0.0007	0.3026	0.8001	0.0133	0.0009	0.0016	0.2566	0.6233	
		8	-0.0057	-0.0001	0.0010	0.0000	0.6167	-0.0211	0.0010	0.0026													
		9																					
772	6	7	0.602	3.10	-0.00	0.1087	1.01	-0.89	-1.01	1.02	0.3328	0.6253	0.0075	0.0001	-0.0008	0.3101	0.5608	0.0219	0.0014	0.0037	0.2625	0.6130	
		8	-0.0039	-0.0002	0.0008	0.0000	1.0416	-0.0308	0.0016	0.0057													
		9																					
772	7	7	0.601	4.14	-0.00	0.2682	1.01	-0.87	-1.02	1.03	0.3209	0.6240	0.0062	0.0003	-0.0007	0.4790	0.7506	0.0304	0.0019	0.0048	0.2673	0.6130	
		8	-0.0027	-0.0002	0.0009	0.0000	1.4488	-0.0397	0.0021	0.0037													
		9																					
772	8	7	0.601	-0.03	-0.00	0.2694	1.01	-0.94	-1.02	0.97	-0.1547	0.9410	0.0064	0.0003	-0.0009	0.3331	0.7956	-0.0016	0.0000	0.0006	0.1706	0.6553	
		8	-0.0044	-0.0002	0.0010	0.0000	1.0292	-0.0042	0.0001	0.0006													
		9																					
773	1	7	0.601	-3.10	-0.00	0.2182	2.08	-0.10	-2.12	0.07	0.2608	0.6196	0.0115	0.0005	-0.0012	0.1517	0.6452	-0.0223	-0.0014	-0.0027	0.3341	0.6280	
		8	-0.0134	-0.0004	0.0014	0.0000	0.4492	-0.0219	0.0014	0.0027													
		9																					
773	2	7	0.601	-0.04	-0.00	0.2396	2.08	-0.05	-2.11	0.02	0.6928	0.5476	0.0128	0.0006	-0.0016	0.2246	0.6487	0.0021	0.0002	0.0004	0.7464	-0.6219	
		8	-0.0116	-0.0005	0.0013	0.0000	0.5914	-0.0032	0.0004	0.0004													
		9																					
773	3	7	0.602	0.46	-0.00	0.3222	2.08	-0.04	-2.11	0.02	0.4482	0.6485	0.0118	0.0007	-0.0014	0.2720	0.7335	0.0057	0.0005	0.0001	0.1351	0.6991	
		8	-0.0106	-0.0005	0.0017	0.0000	0.6910	-0.0040	0.0005	0.0005													
		9																					

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																																	
773	4	7	0.601	0.99	-0.00	0.2171	2.08	-0.03	-2.12	-0.01	0.3715	0.6218	8	0.0131	0.0005	0.0016	0.2171	2.08	-0.03	-2.12	-0.01	0.3715	0.6218	9	-0.0127	-0.0004	-0.0012	0.1768	0.4965	0.0078	0.0002	0.0007	0.0009	0.1787	0.6237
773	5	7	0.602	2.03	-0.00	0.3047	2.08	-0.01	-2.11	-0.00	0.3340	0.6243	8	0.0123	0.0007	0.0017	0.3047	2.08	-0.01	-2.11	-0.00	0.3340	0.6243	9	-0.0102	-0.0005	-0.0014	0.2875	0.7160	0.0185	0.0007	0.0023	0.0019	0.2396	0.6081
773	6	7	0.601	3.08	-0.00	0.2139	2.08	0.00	-2.12	0.00	0.3181	0.6250	8	0.0133	0.0005	0.0016	0.2139	2.08	0.00	-2.12	0.00	0.3181	0.6250	9	-0.0110	-0.0005	-0.0013	0.2311	0.6090	0.0241	0.0012	0.0029	0.0029	0.2619	0.6175
773	7	7	0.601	4.12	0.00	0.2632	2.08	0.01	-2.11	0.02	0.3065	0.6148	8	0.0121	0.0006	0.0015	0.2632	2.08	0.01	-2.11	0.02	0.3065	0.6148	9	-0.0103	-0.0004	-0.0012	0.2255	0.6119	0.0337	0.0017	0.0044	0.0041	0.2656	0.6133
773	8	7	0.601	-0.01	-0.00	0.2744	2.08	-0.05	-2.12	-0.03	0.6105	0.5096	8	0.0123	0.0006	0.0017	0.2744	2.08	-0.05	-2.12	-0.03	0.6105	0.5096	9	-0.0110	-0.0005	-0.0014	0.2631	0.6690	0.0024	0.0002	0.0002	0.0000	-1.2224	0.9240
774	1	7	0.601	3.10	-0.00	0.2182	2.07	-1.90	-2.11	1.92	0.2685	0.6287	8	0.0100	0.0004	0.0013	0.2182	2.07	-1.90	-2.11	1.92	0.2685	0.6287	9	-0.0117	-0.0004	-0.0012	0.1982	0.5503	-0.0116	0.0008	0.0014	0.0014	0.3637	0.6146
774	2	7	0.601	-0.04	-0.00	0.2272	2.08	-1.84	-2.11	1.96	0.1487	0.7417	8	0.0123	0.0005	0.0015	0.2272	2.08	-1.84	-2.11	1.96	0.1487	0.7417	9	-0.0122	-0.0004	-0.0011	0.1696	0.4830	0.0055	0.0003	0.0011	0.0011	0.2060	0.6570
774	3	7	0.602	0.45	-0.00	0.2789	2.08	-1.83	-2.11	1.97	-0.1985	1.1545	8	0.0115	0.0006	0.0016	0.2789	2.08	-1.83	-2.11	1.97	-0.1985	1.1545	9	-0.0098	-0.0005	-0.0014	0.2910	0.7203	-0.0132	0.0006	0.0017	0.0017	0.2475	0.6664
774	4	7	0.601	1.02	-0.00	0.0866	2.07	-1.82	-2.11	1.98	0.6725	0.5246	8	0.0145	0.0002	0.0013	0.0866	2.07	-1.82	-2.11	1.98	0.6725	0.5246	9	-0.0133	-0.0003	-0.0010	0.1290	0.3988	0.0175	0.0008	0.0021	0.0021	0.2474	0.6253
774	5	7	0.601	2.03	-0.00	0.3930	2.08	-1.80	-2.11	2.00	0.3850	0.6309	8	0.0104	0.0008	0.0016	0.3930	2.08	-1.80	-2.11	2.00	0.3850	0.6309	9	-0.0094	-0.0005	-0.0013	0.2868	0.7371	0.0093	0.0007	0.0032	0.0032	0.2650	0.6302
774	6	7	0.602	3.10	-0.00	0.1676	2.08	-1.79	-2.11	2.02	0.3434	0.6209	8	0.0128	0.0004	0.0014	0.1676	2.08	-1.79	-2.11	2.02	0.3434	0.6209	9	-0.0107	-0.0004	-0.0011	0.1878	0.5329	0.0173	0.0011	0.0021	0.0021	0.2612	0.6184

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																			
774	7	0.601	4.12	-0.00	0.2999	2.08	-1.77	-2.11	2.03	0.3259	0.6221	0.0106	0.0006	0.0015	0.1942	0.7095	0.0258	0.0016	0.0032	0.2649	0.6066
	8	-0.0101	-0.0003	-0.0011	0.1462	0.5577	0.0447	0.0023	0.0054												
774	8	0.601	-0.003	-0.00	0.1762	0.5561	-1.84	-2.12	1.97	0.1578	0.8024	0.0128	0.0003	0.0014	0.1462	0.5561	-0.0049	-0.0001	0.0008	0.2084	0.6382
	9	-0.0123	-0.0004	-0.0012	0.2873	0.4932	0.0098	0.0004	0.0012												
775	1	0.601	-3.10	-0.00	0.2796	5.06	-0.10	-5.06	-0.08	0.2677	0.6284	0.0249	0.0015	0.0033	0.2796	0.6169	-0.0226	-0.0011	0.0026	0.3315	0.6315
	8	-0.0249	0.0014	0.0033	0.2795	0.6660	-0.0209	-0.0011	-0.0028												
775	2	0.601	-0.004	-0.00	0.2842	5.06	-0.05	-5.06	-0.04	0.5826	0.5953	0.0275	0.0015	0.0032	0.2842	0.6145	-0.0004	-0.0001	0.0003	0.2677	0.6315
	9	-0.0265	-0.0015	-0.0032	0.2731	0.6412	0.0028	0.0003	0.0003												
775	3	0.601	0.46	-0.00	0.2731	5.07	-0.04	-5.06	-0.03	0.4131	0.6547	0.0276	0.0015	0.0035	0.2731	0.6364	0.0069	0.0005	0.0009	0.1004	0.6309
	8	-0.0276	0.0015	0.0035	0.3048	0.6488	0.0030	0.0000	0.0003												
775	4	0.601	1.03	-0.00	0.2610	5.06	-0.03	-5.07	-0.02	0.3627	0.6294	0.0279	0.0014	0.0034	0.2610	0.6247	0.0112	0.0008	0.0014	0.2049	0.6583
	9	-0.0266	-0.0014	-0.0031	0.2771	0.5955	0.0070	0.0002	0.0009												
775	5	0.601	2.03	0.00	0.2677	5.07	-0.02	-5.07	-0.01	0.3314	0.6263	0.0279	0.0014	0.0031	0.2677	0.6034	0.0194	0.0012	0.0016	0.3314	0.6096
	8	-0.0279	0.0014	0.0035	0.2775	0.6034	0.0149	0.0006	0.0018												
775	6	0.600	3.08	0.00	0.2610	5.06	0.00	-5.06	0.00	0.3170	0.6250	0.0280	0.0014	0.0034	0.2610	0.6203	0.0280	0.0017	0.0035	0.2579	0.6119
	9	-0.0278	-0.0012	-0.0028	0.2279	0.5108	0.0232	0.0011	0.0028												
775	7	0.601	4.13	-0.00	0.2497	5.06	0.01	-5.06	0.02	0.3070	0.6175	0.0277	0.0013	0.0034	0.2497	0.6178	0.0368	0.0022	0.0045	0.2679	0.6175
	8	-0.0277	-0.0013	-0.0030	0.2755	0.6003	0.0326	0.0017	0.0040												
775	8	0.600	-0.05	-0.00	0.2601	5.06	-0.05	-5.07	-0.03	0.5038	0.5371	0.0279	0.0014	0.0034	0.2601	0.6242	0.0033	0.0003	0.0003	0.4774	0.8001
	9	-0.0277	-0.0014	-0.0031	0.2623	0.5752	-0.0003	-0.0001	-0.0000												
776	1	0.600	-3.10	-0.00	0.3195	5.06	-5.13	-5.06	5.00	0.2743	0.6133	0.0224	0.0014	0.0031	0.3195	0.7018	-0.0475	-0.0026	0.0058	0.2743	0.6133
	8	-0.0224	0.0014	0.0031	0.2772	0.6102	-0.0008	-0.0000	-0.0001												
	9	-0.0252	-0.0013	-0.0030	0.2772	0.6102	-0.0008	-0.0000	-0.0001												

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																					
776	2	7	0.601	-0.003	-0.000	0.2635	5.06	-5.07	-5.06	5.05	0.2585	0.6442	7	0.601	-0.003	-0.000	0.2635	5.06	-5.07	-5.06	5.05	0.2585	0.6442
		6	0.0260	0.0013	0.0032	0.2934	0.6316	-0.0203	-0.0010	-0.0026			6	0.0260	0.0013	0.0032	0.2934	0.6316	-0.0203	-0.0010	-0.0026		
		9	-0.0243	-0.0014	-0.0030		0.6279	0.0237	0.0012	0.0030			9	-0.0243	-0.0014	-0.0030		0.6279	0.0237	0.0012	0.0030		
776	3	7	0.600	0.45	-0.000	0.2269	5.06	-5.06	-5.06	5.05	0.2341	0.6504	7	0.600	0.45	-0.000	0.2269	5.06	-5.06	-5.06	5.05	0.2341	0.6504
		6	0.0270	0.0012	0.0031	0.2692	0.5901	-0.0153	-0.0007	-0.0019			6	0.0270	0.0012	0.0031	0.2692	0.5901	-0.0153	-0.0007	-0.0019		
		9	-0.0254	-0.0013	-0.0029		0.5800	0.0291	0.0014	0.0036			9	-0.0254	-0.0013	-0.0029		0.5800	0.0291	0.0014	0.0036		
776	4	7	0.600	1.02	-0.000	0.3097	5.06	-5.05	-5.06	5.06	0.2394	0.7051	7	0.600	1.02	-0.000	0.3097	5.06	-5.05	-5.06	5.06	0.2394	0.7051
		6	0.0251	0.0015	0.0034	0.2729	0.6763	-0.0111	-0.0005	-0.0015			6	0.0251	0.0015	0.0034	0.2729	0.6763	-0.0111	-0.0005	-0.0015		
		9	-0.0245	-0.0013	-0.0029		0.5948	0.0331	0.0017	0.0042			9	-0.0245	-0.0013	-0.0029		0.5948	0.0331	0.0017	0.0042		
776	5	7	0.601	2.05	-0.000	0.3145	5.06	-5.03	-5.05	5.07	0.0984	0.9088	7	0.601	2.05	-0.000	0.3145	5.06	-5.03	-5.05	5.07	0.0984	0.9088
		6	0.0253	0.0015	0.0034	0.3160	0.6814	-0.0030	-0.0000	-0.0005			6	0.0253	0.0015	0.0034	0.3160	0.6814	-0.0030	-0.0000	-0.0005		
		9	-0.0224	-0.0014	-0.0030		0.6740	0.0422	0.0022	0.0052			9	-0.0224	-0.0014	-0.0030		0.6740	0.0422	0.0022	0.0052		
776	6	7	0.601	3.09	-0.000	0.2539	5.06	-5.02	-5.06	5.08	0.5129	0.4827	7	0.601	3.09	-0.000	0.2539	5.06	-5.02	-5.06	5.08	0.5129	0.4827
		6	0.0265	0.0013	0.0032	0.2777	0.6158	0.0032	0.0003	0.0003			6	0.0265	0.0013	0.0032	0.2777	0.6158	0.0032	0.0003	0.0003		
		9	-0.0234	-0.0013	-0.0028		0.6054	0.0499	0.0027	0.0060			9	-0.0234	-0.0013	-0.0028		0.6054	0.0499	0.0027	0.0060		
776	7	7	0.601	4.11	-0.000	0.2632	5.05	-5.00	-5.05	5.11	0.3680	0.5886	7	0.601	4.11	-0.000	0.2632	5.05	-5.00	-5.05	5.11	0.3680	0.5886
		6	0.0257	0.0013	0.0032	0.2758	0.6292	0.0103	0.0007	0.0012			6	0.0257	0.0013	0.0032	0.2758	0.6292	0.0103	0.0007	0.0012		
		9	-0.0226	-0.0012	-0.0027		0.6040	0.0568	0.0031	0.0065			9	-0.0226	-0.0012	-0.0027		0.6040	0.0568	0.0031	0.0065		
776	8	7	0.601	-0.05	-0.000	0.2748	5.06	-5.07	-5.06	5.04	0.2630	0.6510	7	0.601	-0.05	-0.000	0.2748	5.06	-5.07	-5.06	5.04	0.2630	0.6510
		6	0.0256	0.0014	0.0032	0.2985	0.6422	-0.0202	-0.0010	-0.0026			6	0.0256	0.0014	0.0032	0.2985	0.6422	-0.0202	-0.0010	-0.0026		
		9	-0.0240	-0.0014	-0.0030		0.6375	0.0243	0.0012	0.0031			9	-0.0240	-0.0014	-0.0030		0.6375	0.0243	0.0012	0.0031		
77A	3	7	1.252	-3.17	-0.000	-0.3532	-0.02	-3.01	0.05	-3.03	0.1045	0.6668	7	1.252	-3.17	-0.000	-0.3532	-0.02	-3.01	0.05	-3.03	0.1045	0.6668
		6	0.0016	-0.0001	0.0003	0.7134	1.0789	-0.0455	-0.0009	-0.0051			6	0.0016	-0.0001	0.0003	0.7134	1.0789	-0.0455	-0.0009	-0.0051		
		9	-0.0008	0.0001	-0.0002		1.5664	-0.0443	-0.0012	0.0059			9	-0.0008	0.0001	-0.0002		1.5664	-0.0443	-0.0012	0.0059		
77A	4	7	1.252	-0.05	-0.000	-0.2356	-0.02	-2.96	0.05	-2.99	0.1165	0.7015	7	1.252	-0.05	-0.000	-0.2356	-0.02	-2.96	0.05	-2.99	0.1165	0.7015
		6	0.0036	-0.0001	0.0005	0.7821	0.7471	-0.0138	-0.0003	-0.0019			6	0.0036	-0.0001	0.0005	0.7821	0.7471	-0.0138	-0.0003	-0.0019		
		9	-0.0002	0.0001	-0.0002		5.4860	-0.0147	-0.0007	0.0020			9	-0.0002	0.0001	-0.0002		5.4860	-0.0147	-0.0007	0.0020		
77A	5	7	1.251	1.00	-0.000	-0.1702	-0.01	-2.94	0.06	-2.97	0.1115	0.7111	7	1.251	1.00	-0.000	-0.1702	-0.01	-2.94	0.06	-2.97	0.1115	0.7111
		6	0.0033	-0.0001	0.0005	0.6200	0.8450	-0.0036	-0.0000	-0.0006			6	0.0033	-0.0001	0.0005	0.6200	0.8450	-0.0036	-0.0000	-0.0006		
		9	-0.0006	0.0002	-0.0001		0.9355	-0.0051	-0.0004	0.0007			9	-0.0006	0.0002	-0.0001		0.9355	-0.0051	-0.0004	0.0007		

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																			
778	6	7	1.252	3.12	-0.0004	-0.0975	-0.02	-2.91	0.06	-2.94	0.1690	0.6309	0.0026	-0.0001	0.0004	0.7732	0.0130	0.0004	0.0016	0.0376	0.6686
		8	0.0012	0.0000	0.0001	0.6168	0.0108	0.0000	0.0000	0.0000	0.0014										
		9	0.0015	6.26	0.0001	0.4547	0.0413	0.06	0.06	0.0054	0.1260	0.6561	0.0015	0.0002	0.0001	0.3795	0.0415	0.0008	0.0054	0.1015	0.6526
778	7	7	1.251	9.39	-0.0002	-0.3713	-0.02	-2.81	0.06	-2.86	0.1140	0.6464	0.0007	0.0001	0.0002	0.8197	0.0681	0.0013	0.0087	0.1016	0.6442
		8	0.0012	0.0000	0.0001	1.0582	0.0681	0.06	0.06	0.0088											
		9	0.0007	0.0001	0.0001	0.3061	0.0905	0.06	0.06	0.0015	0.1025	0.6305	0.0006	0.0002	0.0001	2.6290	0.0905	0.0016	0.0113	0.0921	0.6293
778	8	7	1.251	12.61	-0.0001	-0.2553	-0.02	-2.78	0.06	-2.84	0.1286	0.7291	0.0002	0.0001	0.0001	0.9484	0.0900	0.0018	0.0113	0.1025	0.6305
		8	0.0027	0.0001	0.0005	1.6467	0.0905	0.06	0.06	0.0016											
		9	0.0011	0.0002	0.0001	0.2553	0.0905	0.06	0.06	0.0016	0.1286	0.7291	0.0001	0.0002	0.0001	0.6733	0.0905	0.0016	0.0113	0.1025	0.6305
779	1	7	1.251	-3.16	-0.0002	-0.5374	-0.02	-2.96	0.05	-1.05	0.1176	0.6798	0.0017	0.0001	0.0002	0.8166	0.0335	0.0007	0.0045	0.1176	0.6798
		8	0.0017	0.0001	0.0002	0.8522	0.0335	0.05	0.05	0.0045											
		9	0.0010	0.0001	0.0002	1.3383	0.0335	0.05	0.05	0.0045	0.1176	0.6798	0.0001	0.0002	0.0001	1.00	0.0335	0.0007	0.0045	0.1176	0.6798
779	2	7	1.250	-0.05	-0.0003	-0.3694	-0.02	-2.95	0.06	-1.01	0.0951	0.8816	0.0032	0.0001	0.0003	0.5967	0.0030	0.0004	0.0005	0.0951	0.8816
		8	0.0032	0.0002	0.0001	1.2841	0.0030	0.06	0.06	0.0005											
		9	0.0009	0.0002	0.0001	0.2558	0.0030	0.06	0.06	0.0005	0.1176	0.6798	0.0001	0.0002	0.0001	1.0148	0.0038	0.0004	0.0005	0.0951	0.8816
779	3	7	1.251	1.00	-0.0004	-0.2558	-0.02	-2.93	0.06	-1.00	0.1176	0.6798	0.0031	0.0001	0.0004	0.6972	0.0057	0.0002	0.0007	0.1176	0.6798
		8	0.0031	0.0001	0.0004	1.7883	0.0057	0.06	0.06	0.0007											
		9	0.0006	0.0002	0.0001	0.2558	0.0057	0.06	0.06	0.0007	0.1176	0.6798	0.0001	0.0002	0.0001	1.1717	0.0040	0.0001	0.0005	0.0951	0.8816
779	4	7	1.251	3.12	-0.0003	-0.1595	-0.02	-2.90	0.06	-0.96	0.1348	0.6520	0.0024	0.0002	0.0003	0.7537	0.0249	0.0006	0.0032	0.1348	0.6520
		8	0.0024	0.0000	0.0003	2.1281	0.0249	0.06	0.06	0.0032											
		9	0.0001	0.0002	0.0000	0.3844	0.0249	0.06	0.06	0.0032	0.1348	0.6520	0.0001	0.0002	0.0003	1.1281	0.0222	0.0003	0.0028	0.1348	0.6520
779	5	7	1.251	6.28	-0.0002	-0.3844	-0.02	-2.85	0.06	-0.91	0.1154	0.6574	0.0012	0.0002	0.0000	0.8351	0.0530	0.0012	0.0069	0.1154	0.6574
		8	0.0012	0.0000	0.0002	0.7746	0.0530	0.06	0.06	0.0069											
		9	0.0017	0.0002	0.0000	0.3300	0.0530	0.06	0.06	0.0069	0.1154	0.6574	0.0001	0.0002	0.0000	0.2549	0.0530	0.0010	0.0069	0.1154	0.6574
779	6	7	1.251	9.41	-0.0002	-0.3300	-0.02	-2.81	0.06	-0.88	0.1063	0.6424	0.0014	0.0002	0.0002	0.6872	0.0785	0.0016	0.0100	0.1063	0.6424
		8	0.0014	0.0000	0.0002	0.5692	0.0785	0.06	0.06	0.0100											
		9	0.0001	0.0002	0.0000	0.2549	0.0785	0.06	0.06	0.0100	0.1063	0.6424	0.0001	0.0002	0.0000	0.5692	0.0785	0.0015	0.0100	0.1063	0.6424

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key														
779	7	1.250	12.63	-0.000	-0.000	-0.5144	-0.002	-0.001	-0.002	-0.02	-0.002	-0.092	0.06	-0.05	0.0962	0.6265
	8	0.0004	0.0002	0.0001	0.0000	3.1644	0.0000	0.0000	0.0000	0.5810	0.0000	0.0993	0.0017	0.0123	0.0882	0.6206
	9	-0.0004	0.0002	-0.0001	-0.0000	-0.1644	-0.0001	-0.0001	-0.0001	-0.9849	-0.0001	0.0993	0.0017	0.0123	0.0882	0.6206
779	8	1.251	-0.003	-0.000	-0.000	-0.2934	-0.000	-0.000	-0.000	-0.02	-0.000	-0.095	0.06	-1.00	0.1299	0.9034
	8	0.0031	0.0001	0.0001	0.0001	0.7434	0.0001	0.0001	0.0001	0.7434	0.0001	-0.031	-0.0004	-0.0005	0.4842	0.7362
	9	-0.0008	0.0002	-0.0001	-0.0001	-1.6162	-0.0001	-0.0001	-0.0001	-0.779	-0.0001	-0.044	-0.0004	-0.0006	0.4842	0.7362
780	1	1.251	-3.13	-0.000	-0.000	-0.5008	-0.000	-0.000	-0.000	-0.02	-0.000	-0.11	0.05	-0.07	0.1284	0.6834
	8	0.0017	0.0001	0.0002	0.0002	0.7843	0.0002	0.0002	0.0002	0.7843	0.0002	-0.274	-0.0007	-0.0037	0.1719	0.6722
	9	-0.0012	0.0002	-0.0002	-0.0002	-1.1510	-0.0002	-0.0002	-0.0002	-1.1510	-0.0002	-0.269	-0.0009	-0.0036	0.1719	0.6722
780	2	1.251	-0.003	-0.000	-0.000	-0.3650	-0.000	-0.000	-0.000	-0.02	-0.000	-0.06	0.06	-0.05	0.3543	0.4950
	8	0.0034	0.0002	0.0003	0.0003	0.5537	0.0003	0.0003	0.0003	0.5537	0.0003	0.012	0.0000	0.0001	-2.1175	0.8062
	9	-0.0009	0.0002	-0.0001	-0.0001	-1.4252	-0.0001	-0.0001	-0.0001	-1.0199	-0.0001	0.005	-0.0002	0.0000	-2.1175	0.8062
780	3	1.250	1.02	-0.000	-0.000	-0.2397	-0.000	-0.000	-0.000	-0.02	-0.000	-0.04	0.05	-0.03	0.1532	0.6519
	8	0.0033	0.0001	0.0004	0.0004	0.6620	0.0004	0.0004	0.0004	0.6620	0.0004	0.109	0.0003	0.0014	0.0232	0.6973
	9	-0.0009	0.0002	-0.0001	-0.0001	-1.3495	-0.0001	-0.0001	-0.0001	-0.7248	-0.0001	0.088	0.0000	0.0012	0.0232	0.6973
780	4	1.250	3.14	-0.000	-0.000	-0.1538	-0.000	-0.000	-0.000	-0.02	-0.000	-0.01	0.06	-0.00	0.1248	0.6589
	8	0.0026	0.0002	0.0003	0.0003	0.6781	0.0003	0.0003	0.0003	0.6781	0.0003	0.308	0.0007	0.0040	0.0950	0.6589
	9	-0.0006	0.0002	-0.0000	-0.0000	-1.6537	-0.0000	-0.0000	-0.0000	-0.6687	-0.0000	0.283	0.0005	0.0037	0.0950	0.6589
780	5	1.251	6.28	-0.000	-0.000	-0.4104	-0.000	-0.000	-0.000	-0.02	-0.000	0.02	0.05	0.04	0.1121	0.6574
	8	0.0013	0.0001	0.0001	0.0001	0.6796	0.0001	0.0001	0.0001	0.6796	0.0001	0.583	0.0013	0.0076	0.1000	0.6574
	9	-0.0017	0.0003	-0.0001	-0.0001	-0.8797	-0.0001	-0.0001	-0.0001	-0.3731	-0.0001	0.586	0.0011	0.0076	0.1000	0.6574
780	6	1.250	9.42	-0.000	-0.000	-0.3601	-0.000	-0.000	-0.000	-0.02	-0.000	0.07	0.06	0.07	0.1045	0.6389
	8	0.0014	0.0001	0.0002	0.0002	0.7089	0.0002	0.0002	0.0002	0.7089	0.0002	0.829	0.0017	0.0106	0.0967	0.6389
	9	-0.0003	0.0002	-0.0001	-0.0001	-2.9293	-0.0001	-0.0001	-0.0001	-1.5180	-0.0001	0.829	0.0016	0.0105	0.0967	0.6389
780	7	1.250	12.64	-0.000	-0.000	-0.4612	-0.000	-0.000	-0.000	-0.02	-0.000	0.10	0.05	0.10	0.0932	0.6229
	8	0.0009	0.0002	0.0001	0.0001	0.5557	0.0001	0.0001	0.0001	0.5557	0.0001	1.032	0.0019	0.0128	0.0672	0.6176
	9	-0.0006	0.0002	-0.0001	-0.0001	-1.8392	-0.0001	-0.0001	-0.0001	-0.8929	-0.0001	1.033	0.0018	0.0127	0.0672	0.6176
780	8	1.250	-0.002	-0.000	-0.000	-0.3256	-0.000	-0.000	-0.000	-0.02	-0.000	-0.05	0.05	-0.05	0.2079	0.3503
	8	0.0032	0.0002	0.0004	0.0004	0.6412	0.0004	0.0004	0.0004	0.6412	0.0004	0.014	0.0000	0.0001	-2.0206	0.3503
	9	-0.0009	0.0002	-0.0001	-0.0001	-1.5672	-0.0001	-0.0001	-0.0001	-1.0465	-0.0001	0.006	-0.0002	0.0000	-2.0206	0.3503
781	1	1.251	-3.11	-0.000	-0.000	-0.5173	-0.000	-0.000	-0.000	-0.02	-0.000	0.96	0.05	0.93	0.1346	0.6749
	8	0.0016	0.0001	0.0002	0.0002	0.8452	0.0002	0.0002	0.0002	0.8452	0.0002	0.209	0.0005	0.0028	0.1979	0.6749
	9	-0.0016	0.0002	-0.0002	-0.0002	-0.7334	-0.0002	-0.0002	-0.0002	-0.8391	-0.0002	-0.205	-0.0008	-0.0027	0.1979	0.6749

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key															
782	5	1.251	6.32	-0.001	-0.4620	-0.02	3.12	0.06	3.15	0.1059	0.6518	0.0015	0.0014	0.0015	0.0096	0.0957	0.6428
	8	0.0015	-0.0003	0.0001	-0.5533	0.7248	0.0746	0.0015	0.0096	0.0957	0.6428	0.0015	0.0014	0.0015	0.0096	0.0957	0.6428
	9	0.0010	0.0003	0.0001	1.8362	0.7248	0.0752	0.0014	0.0096	0.0957	0.6428	0.0014	0.0014	0.0014	0.0096	0.0957	0.6428
782	6	1.250	9.45	-0.001	-0.3770	-0.02	3.15	0.06	3.18	0.0962	0.6307	0.0013	0.0017	0.0018	0.0122	0.0903	0.6263
	8	0.0013	-0.0001	0.0001	5.9405	-3.4566	0.0972	0.0017	0.0122	0.0903	0.6263	0.0013	0.0017	0.0018	0.0122	0.0903	0.6263
	9	0.0001	0.0002	-0.0001	-0.3770	3.4566	0.0972	0.0017	0.0122	0.0903	0.6263	0.0001	0.0002	0.0001	0.0122	0.0903	0.6263
782	7	1.251	12.66	-0.001	-0.3733	-0.02	3.18	0.05	3.21	0.0922	0.6103	0.0007	0.0002	0.0021	0.0139	0.0843	0.6037
	8	0.0007	-0.0000	0.0001	11.1859	-5.5773	0.1140	0.0019	0.0139	0.0843	0.6037	0.0007	0.0002	0.0021	0.0139	0.0843	0.6037
	9	0.0001	0.0002	-0.0001	-0.3733	5.5773	0.1148	0.0019	0.0138	0.0843	0.6037	0.0001	0.0002	0.0001	0.0138	0.0843	0.6037
782	8	1.251	-0.03	-0.001	-0.2643	-0.02	3.02	0.06	3.08	0.1342	0.6623	0.0025	0.0004	0.0004	0.0024	0.0773	0.6668
	8	0.0025	-0.0001	0.0004	-2.3431	2.8287	0.0181	0.0004	0.0024	0.0773	0.6668	0.0025	0.0004	0.0004	0.0024	0.0773	0.6668
	9	-0.0005	0.0002	-0.0003	2.3431	-2.8287	0.0162	0.0004	0.0021	0.0773	0.6668	-0.0005	0.0002	0.0003	0.0021	0.0773	0.6668
783	4	1.251	3.08	-0.001	-0.7491	-0.02	6.00	0.05	6.01	0.2871	0.7991	0.0012	0.0002	0.0003	0.0008	0.0304	0.7991
	6	0.0012	-0.0001	0.0002	0.7491	1.0807	0.0055	0.0002	0.0008	0.0304	0.7991	0.0012	0.0002	0.0003	0.0008	0.0304	0.7991
	9	-0.0018	0.0001	-0.0002	-0.7491	0.7610	0.0059	0.0002	0.0009	0.0304	0.7991	-0.0018	0.0001	0.0002	0.0009	0.0304	0.7991
783	5	1.250	0.00	-0.001	-0.3099	-0.02	6.05	0.05	6.06	0.1234	0.6909	0.0027	0.0004	0.0006	0.0050	0.0925	0.6856
	8	0.0027	-0.0001	0.0004	0.3099	0.783	0.0367	0.0006	0.0050	0.0925	0.6856	0.0027	0.0004	0.0006	0.0050	0.0925	0.6856
	9	-0.0013	0.0002	-0.0001	-0.3099	0.6407	0.0355	0.0006	0.0048	0.0925	0.6856	-0.0013	0.0002	0.0001	0.0048	0.0925	0.6856
783	6	1.249	1.08	-0.001	-0.2018	-0.02	6.07	0.05	6.07	0.1159	0.6851	0.0028	0.0004	0.0008	0.0065	0.0951	0.6763
	8	0.0028	-0.0001	0.0004	0.2018	0.8026	0.0475	0.0008	0.0065	0.0951	0.6763	0.0028	0.0004	0.0008	0.0065	0.0951	0.6763
	9	-0.0010	0.0002	-0.0001	-0.2018	0.6058	0.0459	0.0008	0.0062	0.0951	0.6763	-0.0010	0.0002	0.0001	0.0062	0.0951	0.6763
783	7	1.249	3.19	-0.001	-0.0107	-0.02	6.10	0.06	6.10	0.1118	0.6653	0.0020	0.0014	0.0012	0.0087	0.0963	0.6582
	8	0.0020	0.0000	0.0003	0.0107	0.9628	0.0654	0.0012	0.0087	0.0963	0.6582	0.0020	0.0014	0.0012	0.0087	0.0963	0.6582
	9	-0.0003	0.0003	0.0000	-4.1935	-0.1692	0.0645	0.0012	0.0084	0.0963	0.6582	-0.0003	0.0003	0.0000	0.0084	0.0963	0.6582
783	8	1.254	6.33	-0.001	-0.5447	-0.02	6.14	0.06	6.13	0.1027	0.6429	0.0006	0.0016	0.0016	0.0113	0.0911	0.6331
	8	0.0006	-0.0000	0.0002	0.5447	1.7996	0.0883	0.0016	0.0113	0.0911	0.6331	0.0006	0.0016	0.0016	0.0113	0.0911	0.6331
	9	0.0012	0.0003	0.0001	-1.3288	0.5086	0.0896	0.0016	0.0113	0.0911	0.6331	0.0012	0.0003	0.0001	0.0113	0.0911	0.6331
783	9	1.251	9.47	-0.001	-0.4244	-0.01	6.17	0.06	6.15	0.0952	0.6184	0.0010	0.0020	0.0018	0.0134	0.0855	0.6106
	8	0.0010	-0.0000	0.0001	0.4244	0.9564	0.1084	0.0018	0.0134	0.0855	0.6106	0.0010	0.0020	0.0018	0.0134	0.0855	0.6106
	9	-0.0002	0.0002	-0.0000	-4.6365	0.9835	0.1093	0.0018	0.0133	0.0855	0.6106	-0.0002	0.0000	0.0000	0.0133	0.0855	0.6106
783	10	1.250	12.64	-0.001	-0.5594	-0.01	6.19	0.06	6.17	0.0923	0.5996	0.0006	0.0022	0.0021	0.0144	0.0881	0.5996
	8	0.0006	-0.0000	0.0001	0.5594	1.0008	0.1201	0.0022	0.0144	0.0881	0.5996	0.0006	0.0022	0.0021	0.0144	0.0881	0.5996
	9	-0.0008	0.0002	-0.0000	-1.6459	0.2050	0.1221	0.0021	0.0143	0.0881	0.5996	-0.0008	0.0000	0.0000	0.0143	0.0881	0.5996

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key													
785	3	7	1.250	1.13	-0.000	-0.000	-0.000	-0.1076	-0.8669	15.15	0.0017	0.0962	0.6400		
		8	0.0023	-0.0000	0.0004	0.0001	-1.5013	1.1051	0.0930	0.0016	0.0889	0.6380			
		9	-0.0000	0.0002	-0.0001	-0.0001	-1.5013	1.1051	0.0923	0.0016	0.0889	0.6380			
785	4	7	1.255	3.22	-0.000	-0.000	-0.0528	-0.7824	15.18	0.0020	0.0970	0.6220			
		8	0.0019	-0.0000	0.0002	-0.0000	-4.7659	1.1501	0.1050	0.0018	0.0887	0.6153			
		9	-0.0003	0.0003	-0.0000	-0.0000	-4.7659	1.1501	0.1050	0.0018	0.0887	0.6153			
785	5	7	1.254	6.37	-0.000	-0.000	-0.4968	-0.6369	15.19	0.0022	0.0974	0.5993			
		8	0.0012	-0.0001	0.0001	-0.0001	1.7171	0.7039	0.1165	0.0023	0.0982	0.5850			
		9	0.0010	0.0003	0.0001	0.0001	1.7171	0.7039	0.1183	0.0023	0.0982	0.5850			
785	6	7	1.249	9.50	-0.000	-0.000	-0.2885	-0.8444	15.21	0.0022	0.0886	0.5908			
		8	0.0009	-0.0000	0.0002	-0.0000	-3.4593	1.0628	0.1287	0.0022	0.0873	0.5794			
		9	-0.0003	0.0002	-0.0000	-0.0000	-3.4593	1.0628	0.1296	0.0022	0.0873	0.5794			
785	7	7	1.250	12.70	-0.000	-0.000	-0.4861	-0.8429	15.21	0.0021	0.0774	0.5853			
		8	0.0007	-0.0000	0.0001	-0.0000	-1.0446	0.8429	0.1347	0.0021	0.0785	0.5694			
		9	-0.0015	0.0003	-0.0000	-0.0000	-1.0446	0.8429	0.1358	0.0021	0.0785	0.5694			
785	8	7	1.251	0.06	-0.000	-0.000	-0.2907	-0.3003	15.14	0.0015	0.0986	0.6484			
		8	0.0023	-0.0001	0.0003	-0.0001	-0.9079	0.7681	0.0846	0.0015	0.0899	0.6467			
		9	-0.0017	0.0003	-0.0001	-0.0001	-0.9079	0.7681	0.0844	0.0015	0.0899	0.6467			
786	1	7	0.899	-3.01	-0.000	-0.000	-0.4724	-0.3458	15.10	0.0027	0.2257	0.6106			
		8	0.0024	-0.0002	0.0003	-0.0001	-0.4726	0.7454	0.0601	0.0025	0.2092	0.6150			
		9	-0.0028	0.0002	-0.0001	-0.0001	-0.4726	0.7458	0.0614	0.0025	0.2092	0.6150			
786	2	7	0.900	0.03	-0.000	-0.000	-0.1722	-1.3777	15.12	0.0032	0.2018	0.5755			
		8	0.0024	-0.0000	0.0003	-0.0002	-0.9925	0.7797	0.0793	0.0030	0.1890	0.5710			
		9	-0.0009	0.0001	-0.0002	-0.0002	-0.9925	1.3777	0.0804	0.0030	0.1890	0.5710			
786	3	7	0.900	1.09	-0.000	-0.000	-0.0819	-0.3273	15.13	0.0031	0.1841	0.5717			
		8	0.0022	-0.0000	0.0003	-0.0002	-0.9978	0.7386	0.0858	0.0031	0.1798	0.5680			
		9	-0.0008	0.0001	-0.0002	-0.0002	-0.9978	1.3273	0.0863	0.0031	0.1798	0.5680			
786	4	7	0.900	3.15	-0.000	-0.000	0.2465	1.1975	15.14	0.0029	0.1601	0.5640			
		8	0.0012	0.0000	0.0003	-0.0001	-1.0708	0.8512	0.0936	0.0029	0.1560	0.5575			
		9	-0.0011	0.0002	-0.0001	-0.0001	-1.0708	0.8512	0.0947	0.0029	0.1560	0.5575			
786	5	7	0.900	6.25	-0.000	-0.000	-0.8122	-0.9675	15.13	0.0024	0.1223	0.5668			
		8	0.0009	-0.0001	0.0001	-0.0000	-2.5829	0.1366	0.1011	0.0025	0.1220	0.5578			
		9	-0.0006	0.0003	-0.0000	-0.0000	-2.5829	0.1366	0.1035	0.0025	0.1220	0.5578			

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key															
786	6	7	0.899	9.34	-0.0001	-0.0001	-1.3529	-0.03	15.14	0.05	15.12	0.1135	0.5642					
		8	0.0006	-0.0001	0.0001	0.0001	1.5469	-0.03	0.1109	0.0025	0.0125	0.1151	0.5535					
		9	-0.0008	0.0002	-0.0001	-0.0001	0.6277	0.1126	0.0025	0.0124								
786	7	7	0.901	12.48	0.0002	0.0002	-0.9208	-0.04	15.14	0.06	15.11	0.0947	0.5679					
		8	0.0008	-0.0001	0.0002	0.0002	1.1485	0.1190	0.0022	0.0135								
		9	-0.0019	0.0002	-0.0001	-0.0001	0.4179	0.1197	0.0022	0.0134								
786	8	7	0.897	0.03	-0.0003	-0.0003	-0.2869	-0.03	15.13	0.06	15.10	0.1994	0.5709					
		8	0.0028	-0.0001	0.0003	0.0003	0.5868	0.0797	0.0031	0.0091								
		9	-0.0018	0.0002	-0.0001	-0.0001	0.4505	0.0805	0.0030	0.0092								
787	1	7	0.900	-3.03	-0.0002	-0.0002	-0.4576	-0.04	9.08	0.05	9.01	0.2623	0.6808					
		8	0.0024	-0.0002	0.0003	0.0003	0.7009	0.269	0.0014	0.0036								
		9	-0.0024	0.0002	-0.0002	-0.0002	0.5157	0.0276	0.0011	0.0037								
787	2	7	0.901	0.00	-0.0004	-0.0004	-0.2761	-0.03	9.13	0.05	9.05	0.2292	0.6149					
		8	0.0038	-0.0002	0.0004	0.0004	0.6377	0.351	0.0025	0.0067								
		9	-0.0013	0.0001	-0.0002	-0.0002	0.8296	0.0545	0.0022	0.0068								
787	3	7	0.899	1.07	-0.0002	-0.0002	-0.1884	-0.02	9.14	0.05	9.06	0.2201	0.5985					
		8	0.0034	-0.0001	0.0004	0.0004	0.6461	0.629	0.0027	0.0075								
		9	-0.0009	0.0001	-0.0002	-0.0002	1.2134	0.0622	0.0025	0.0075								
787	4	7	0.899	3.14	-0.0003	-0.0003	10.2212	-0.02	9.16	0.06	9.09	0.2083	0.5744					
		8	0.0017	0.0000	0.0002	0.0002	0.6716	0.0763	0.0031	0.0087								
		9	0.0000	0.0001	-0.0003	-0.0003	22.4817	0.0770	0.0029	0.0088								
787	5	7	0.902	6.24	-0.0001	-0.0001	-0.7172	-0.03	9.17	0.06	9.09	0.1677	0.5639					
		8	0.0010	-0.0001	0.0001	0.0001	0.5887	0.0911	0.0030	0.0102								
		9	-0.0007	0.0003	-0.0000	-0.0000	-0.3102	0.0935	0.0030	0.0103								
787	6	7	0.900	9.32	-0.0002	-0.0002	-1.3801	-0.03	9.17	0.05	9.09	0.1242	0.5646					
		8	0.0007	-0.0002	0.0001	0.0001	1.0405	0.1009	0.0025	0.0113								
		9	-0.0005	0.0001	-0.0001	-0.0001	1.2644	0.1041	0.0025	0.0115								
787	7	7	0.901	12.47	0.0000	0.0000	1.2483	-0.03	9.18	0.05	9.10	0.1132	0.5646					
		8	0.0003	-0.0000	0.0001	0.0001	-2.2957	0.1109	0.0025	0.0125								
		9	-0.0003	0.0000	-0.0003	-0.0003	4.3936	0.1143	0.0026	0.0125								
787	8	7	0.898	0.03	-0.0005	-0.0005	-0.2555	-0.04	9.13	0.05	9.05	0.2290	0.6142					
		8	0.0031	-0.0001	0.0005	0.0005	0.6102	0.0557	0.0025	0.0068								
		9	-0.0015	0.0002	-0.0001	-0.0001	0.5536	0.0556	0.0022	0.0068								

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key															
78A	1	7	0.901	-3.06	-0.003	-0.0003	-0.4266	-0.7357	-0.04	6.02	0.05	6.00	0.3531	0.6818			
	8	8	0.0023	-0.0001	0.0003	0.0002	0.4448	0.5833	0.0082	0.0005	0.0011	0.2305	0.6936				
	9	9	-0.0025	0.0002	-0.0002	-0.0002	-0.4448	0.5833	0.0079	0.0003	0.0011	0.2305	0.6936				
78A	2	7	0.899	-0.001	-0.0005	-0.0002	-0.2497	-0.7977	-0.03	6.06	0.05	6.04	0.2307	0.6496			
	8	8	0.0035	-0.0001	0.0005	0.0002	0.7208	0.6860	0.0399	0.0018	0.0051	0.1997	0.6485				
	9	9	-0.0014	0.0002	-0.0002	-0.0002	-0.7208	0.6860	0.0387	0.0015	0.0050	0.1997	0.6485				
78A	3	7	0.900	3.11	-0.0003	-0.0002	0.0561	-0.02	6.10	0.05	6.07	0.2198	0.5971				
	8	8	0.0017	0.0000	0.0003	0.0002	0.9891	0.9780	0.0640	0.0028	0.0076	0.2050	0.5996				
	9	9	-0.0009	0.0001	-0.0002	-0.0002	-0.9891	1.0428	0.0635	0.0026	0.0076	0.2050	0.5996				
78A	4	7	0.899	6.23	-0.0003	-0.0002	-1.0880	1.7211	-0.02	6.12	0.06	6.10	0.1853	0.5736			
	8	8	0.0002	-0.0002	0.0000	0.0000	-5.7554	0.2677	0.0845	0.0031	0.0097	0.1821	0.5661				
	9	9	-0.0002	0.0002	-0.0000	-0.0000	5.7554	0.2677	0.0856	0.0031	0.0097	0.1821	0.5661				
78A	5	7	0.898	9.32	-0.0001	-0.0002	-1.1727	1.2310	-0.03	6.13	0.05	6.10	0.1466	0.5646			
	8	8	0.0007	-0.0001	0.0001	0.0001	-1.2106	0.5573	0.0973	0.0028	0.0109	0.1437	0.5528				
	9	9	-0.0008	0.0002	-0.0000	-0.0000	1.2106	0.5573	0.1000	0.0028	0.0110	0.1437	0.5528				
78A	6	7	0.899	12.46	0.0001	0.0003	1.8555	-1.6145	-0.03	6.12	0.05	6.10	0.1146	0.5667			
	8	8	0.0003	-0.0001	0.0001	0.0003	-0.6789	-1.7636	0.03	6.12	0.05	6.10	0.1194	0.5532			
	9	9	-0.0008	0.0001	-0.0003	-0.0003	0.6789	1.7636	0.03	6.12	0.05	6.10	0.1194	0.5532			
78A	7	7	0.901	-0.0001	-0.0002	-0.0002	-0.2151	-0.7450	-0.02	6.06	0.05	6.04	0.2304	0.6471			
	8	8	0.0037	-0.0001	0.0005	0.0002	0.9925	1.3777	0.02	6.06	0.05	6.04	0.2026	0.6512			
	9	9	-0.0009	0.0001	-0.0002	-0.0002	-0.9925	1.3777	0.02	6.06	0.05	6.04	0.2026	0.6512			
78A	1	7	0.900	-3.07	-0.0003	-0.0003	-0.3598	-0.8115	-0.04	2.97	0.05	3.03	0.1717	0.7118			
	8	8	0.0022	-0.0001	0.0003	0.0003	-0.5400	0.8733	0.04	2.97	0.05	3.03	0.3349	0.7282			
	9	9	-0.0018	0.0002	-0.0003	-0.0003	0.5400	0.8733	0.04	2.97	0.05	3.03	0.3349	0.7282			
78A	2	7	0.900	-0.0001	-0.0002	-0.0002	-0.2320	-0.6984	-0.03	3.02	0.05	3.07	0.2605	0.6649			
	8	8	0.0038	-0.0001	0.0005	0.0002	-1.0114	1.6449	0.03	3.02	0.05	3.07	0.1850	0.6631			
	9	9	-0.0008	0.0001	-0.0002	-0.0002	1.0114	1.6449	0.03	3.02	0.05	3.07	0.1850	0.6631			
78A	3	7	0.899	1.01	-0.0006	-0.0002	-0.0907	-0.9130	-0.03	3.03	0.05	3.08	0.2331	0.6493			
	8	8	0.0033	-0.0000	0.0006	0.0002	-0.8734	1.3112	0.03	3.03	0.05	3.08	0.1926	0.6441			
	9	9	-0.0009	0.0001	-0.0002	-0.0002	0.8734	1.3112	0.03	3.03	0.05	3.08	0.1926	0.6441			
78A	4	7	0.900	3.17	-0.0003	-0.0002	-0.2014	-0.7208	-0.03	3.06	0.05	3.11	0.2238	0.6191			
	8	8	0.0028	-0.0001	0.0003	0.0002	-2.0631	2.4467	0.03	3.06	0.05	3.11	0.1960	0.6267			
	9	9	-0.0008	0.0001	-0.0002	-0.0002	2.0631	2.4467	0.03	3.06	0.05	3.11	0.1960	0.6267			

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key														
789	5	7	8	9	0.899	6.22	-0.000	-0.000	-1.5231	-0.8544	-0.0067	3.09	0.06	3.14	0.2072	0.5811
					0.0001	-0.0000	-0.0000	-0.0000	-3.2194	0.0067	0.0736	0.0718	0.0029	0.0083	0.1929	0.5778
					-0.0004	0.0002	-0.0000	-0.0000					0.0028	0.0085		
789	6	7	8	9	0.899	9.28	-0.000	-0.000	-1.2346	1.1946	0.6200	3.11	0.05	3.16	0.1731	0.5639
					0.0006	-0.0001	0.0001	0.0001	-1.0657	0.6200	0.0923	0.0905	0.0031	0.0102	0.1693	0.5531
					-0.0006	0.0001	-0.0001	-0.0001					0.0031	0.0102		
789	7	7	8	9	0.899	12.46	0.000	0.000	-2.2001	2.1559	0.5546	3.11	0.05	3.16	0.1274	0.5662
					0.0003	-0.0001	0.0001	0.0001	-0.5818	0.5546	0.1065	0.1027	0.0026	0.0116	0.1259	0.5534
					-0.0015	0.0001	-0.0001	-0.0001					0.0026	0.0116		
789	8	7	8	9	0.900	-0.000	-0.000	-0.000	-0.1243	1.0093	0.8156	3.02	0.05	3.07	0.2530	0.6539
					0.0024	-0.0002	0.0005	0.0005	-0.7502	1.0093	0.0189	0.0197	0.0010	0.0025	0.1850	0.6608
					-0.0014	0.0002	-0.0002	-0.0002					0.0007	0.0025		
790	1	7	8	9	0.899	-3.08	-0.000	-0.000	-0.3250	0.8656	-0.0206	0.97	0.05	0.92	0.1943	0.6772
					0.0020	-0.0001	0.0003	0.0003	-0.6176	0.8656	-0.0186	-0.0206	-0.0008	-0.0027	0.2934	0.6889
					-0.0013	0.0001	-0.0003	-0.0003					-0.0010	-0.0025		
790	2	7	8	9	0.899	-0.000	-0.000	-0.000	-0.0638	0.9364	0.0067	1.03	0.06	0.96	0.3915	0.5968
					0.0030	-0.0001	0.0005	0.0005	-0.8235	0.9364	0.0060	0.0067	0.0005	0.0007	0.1175	0.6285
					-0.0011	0.0001	-0.0002	-0.0002					0.0001	0.0007		
790	3	7	8	9	0.900	0.98	-0.000	-0.000	-0.0681	0.9291	0.0175	1.05	0.05	0.98	0.2520	0.6572
					0.0032	-0.0000	0.0006	0.0006	-1.0267	0.9291	0.0155	0.0175	0.0008	0.0023	0.1677	0.6451
					-0.0007	0.0001	-0.0002	-0.0002					0.0008	0.0023		
790	4	7	8	9	0.900	3.10	-0.000	-0.000	-0.2251	0.6687	0.0363	1.08	0.05	1.01	0.2239	0.6405
					0.0032	-0.0001	0.0004	0.0004	-1.0287	0.6687	0.0363	0.0393	0.0017	0.0050	0.1948	0.6460
					-0.0010	0.0002	-0.0001	-0.0001					0.0014	0.0046		
790	5	7	8	9	0.901	6.21	-0.000	-0.000	0.5165	0.7705	0.0640	1.12	0.06	1.04	0.2141	0.5944
					0.0001	-0.0002	0.0000	0.0000	-5.3447	0.7705	0.0640	0.0633	0.0027	0.0075	0.1991	0.5929
					-0.0002	0.0002	-0.0000	-0.0000					0.0025	0.0075		
790	6	7	8	9	0.901	9.31	-0.000	-0.000	-1.7231	1.4486	0.0844	1.14	0.05	1.07	0.1836	0.5726
					0.0002	-0.0000	0.0000	0.0000	-2.2838	1.4486	0.0857	0.0844	0.0031	0.0096	0.1782	0.5622
					-0.0003	0.0001	-0.0001	-0.0001					0.0030	0.0096		
790	7	7	8	9	0.899	12.45	0.000	0.000	-1.0607	0.6370	0.1021	1.15	0.05	1.07	0.1432	0.5663
					0.0011	-0.0002	0.0001	0.0001	-0.8835	0.6370	0.1021	0.0997	0.0028	0.0113	0.1403	0.5547
					-0.0006	0.0001	-0.0002	-0.0002					0.0028	0.0113		

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key														
790	8	7 6 5	0.900 0.0031 -0.00009	-0.0000 0.0000 0.0001	-0.0005 -0.0002 -0.0002	-0.1322 -0.9279	-0.03 0.768 1.3132	1.03 0.067 0.0061	0.05 0.0005 0.0001	0.0008 0.0007	0.3915 0.1333	0.5968 0.6308					
791	1	7 6 5	0.900 0.0022 -0.00021	-3.09 -0.0001 0.0002	-0.0003 -0.0002 -0.0002	-0.3851 -0.5235	-0.04 0.7170 0.6501	-0.11 -0.0272 -0.0254	0.05 -0.0011 -0.0013	-0.06 -0.0036 -0.0034	0.2168 0.2728	0.6664 0.6678					
791	2	7 6 5	0.900 0.0034 -0.00010	-0.04 -0.0001 0.0001	-0.0005 -0.0002 -0.0002	-0.1954 -0.6707	-0.03 0.7240 1.1401	-0.06 0.0017 0.0009	0.05 0.0001 0.0000	-0.01 0.0000 0.0000	0.5185 -0.4740	0.2658 0.1782					
791	3	7 6 5	0.900 0.0039 -0.00006	0.96 -0.0001 0.0001	-0.0005 -0.0002 -0.0002	-0.1777 -1.1395	-0.03 0.6979 2.1538	-0.04 0.0109 0.0089	0.05 0.0006 0.0002	-0.00 0.0014 0.0011	0.2846 0.1470	0.6502 0.6353					
791	4	7 6 5	0.901 0.0026 -0.00016	3.10 -0.0000 0.0002	-0.0005 -0.0000 -0.0000	-0.1175 -0.7748	-0.03 0.9483 0.2754	-0.01 0.0322 0.0301	0.05 0.0014 0.0010	0.01 0.0041 0.0037	0.2231 0.1821	0.6409 0.6254					
791	5	7 6 5	0.900 0.0001 -0.00003	6.19 -0.0000 0.0003	-0.0000 -0.0000 -0.0000	-1.2972 -3.9872	-0.03 -1.1925 -0.3383	7.03 0.589 0.0606	0.06 0.0025 0.0023	0.06 0.0071 0.0073	0.2158 0.1921	0.6036 0.6016					
791	6	7 6 5	0.901 0.0001 -0.00006	9.28 -0.0000 0.0001	-0.0000 -0.0000 -0.0000	-1.9519 -1.4989	-0.02 1.6201 0.5114	0.05 0.0804 0.0821	0.05 0.0030 0.0029	0.09 0.0092 0.0092	0.1903 0.1824	0.5718 0.5658					
791	7	7 6 5	0.900 0.0011 -0.0001A	12.46 -0.0002 0.0001	0.0001 -0.0001 -0.0001	-0.8979 -0.5200	-0.04 0.7932 0.3893	0.07 0.0973 0.0990	0.05 0.0030 0.0029	0.10 0.0109 0.0109	0.1554 0.1475	0.5633 0.5537					
791	8	7 6 5	0.900 0.0034 -0.00012	-0.04 -0.0001 0.0002	-0.0005 -0.0002 -0.0002	-0.1851 -0.8246	-0.02 0.7708 0.9494	-0.06 0.0020 0.0009	0.05 0.0001 0.0000	-0.01 0.0001 0.0000	0.4347 -0.4316	0.2637 0.0562					
792	1	7 6 5	0.902 0.0021 -0.00009	-3.11 -0.0001 0.0000	-0.0003 -0.0003 -0.0003	-0.3639 -0.4767	-0.03 0.754A 1.7213	-3.02 -0.0460 -0.0427	0.05 -0.0019 -0.0022	-3.01 -0.0059 -0.0054	0.2135 0.2593	0.6490 0.6397					
792	2	7 6 5	0.899 0.0030 -0.0000A	-0.06 -0.0000 0.0001	-0.0005 -0.0002 -0.0002	-0.0901 -1.0619	-0.03 0.9154 1.4756	-2.96 -0.0148 -0.0143	0.05 -0.0005 -0.0009	-2.97 -0.0021 -0.0020	0.1937 0.3320	0.7104 0.7141					

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key													
792	3	7	0.899	0.99	-0.000	0.005	-0.0863	-0.003	-2.94	0.05	-2.96	0.3020	0.8972	0.4114	0.8037
		8	0.0035	-0.0000	0.0000	0.0005	0.8057	0.8057	-0.0036	-0.0002	-0.0006				
		9	-0.0008	0.0001	-0.0002	-0.0002	1.4519	1.4519	-0.0062	-0.0005	-0.0010				
792	4	7	0.899	3.07	-0.000	0.004	-0.2775	-0.003	-2.91	0.05	-2.93	0.2359	0.6265	0.1282	0.6084
		8	0.0039	-0.0002	0.0004	0.0004	0.5632	0.5632	0.0132	0.0006	0.0016				
		9	0.0006	0.0000	-0.0003	-0.0003	-2.9761	-2.9761	0.0108	0.0002	0.0013				
792	5	7	0.898	6.15	-0.000	0.000	0.5916	-0.003	-2.96	0.06	-2.89	0.2157	0.6321	0.1873	0.6251
		8	-0.0000	-0.0000	0.0001	0.0001	1.573	1.573	0.0433	0.0018	0.0054				
		9	-0.0009	0.0003	0.0000	0.0000	-1.6391	-0.2216	0.0439	0.0016	0.0054				
792	6	7	0.899	9.27	-0.000	0.000	1.0669	-0.002	-2.82	0.05	-2.85	0.2058	0.5859	0.1900	0.5804
		8	-0.0001	-0.0002	-0.0000	-0.0000	2.2121	2.2121	0.0685	0.0028	0.0080				
		9	-0.0012	0.0002	-0.0000	-0.0000	0.0963	0.0963	0.0695	0.0026	0.0080				
792	7	7	0.900	12.45	-0.000	0.000	-1.1851	-0.003	-2.80	0.05	-2.83	0.1697	0.5674	0.1664	0.5623
		8	0.0008	-0.0002	0.0001	0.0001	0.7586	0.7586	0.0900	0.0030	0.0102				
		9	-0.0018	0.0001	-0.0001	-0.0001	0.4236	0.4236	0.0915	0.0030	0.0103				
792	8	7	0.900	-0.07	-0.000	0.000	-0.1862	-0.002	-2.96	0.05	-2.97	0.1947	0.7144	0.3262	0.7167
		8	0.0034	-0.0001	0.0005	0.0005	0.7290	0.7290	-0.0146	-0.0005	-0.0020				
		9	-0.0010	0.0002	-0.0002	-0.0002	1.0814	1.0814	-0.0143	-0.0009	-0.0020				
793	6	7	0.599	-3.12	-0.000	0.000	-0.4565	-0.002	-3.02	0.05	-3.03	0.2789	0.6300	0.3159	0.6151
		8	0.0015	-0.0001	0.0001	0.0001	0.478	0.478	-0.0389	-0.0021	-0.0049				
		9	-0.0019	0.0000	-0.0002	-0.0002	0.5158	0.5158	-0.0369	-0.0023	-0.0045				
793	7	7	0.601	-0.05	-0.000	0.000	-0.2172	-0.002	-2.96	0.05	-2.98	0.2427	0.6793	0.3860	0.6502
		8	0.0017	0.0000	0.0004	0.0004	1.3356	1.3356	-0.0112	-0.0005	-0.0015				
		9	-0.0011	0.0000	-0.0002	-0.0002	0.9569	0.9569	-0.0113	-0.0008	-0.0014				
793	8	7	0.601	1.00	-0.000	0.000	0.3069	-0.002	-2.94	0.05	-2.96	0.1597	0.9271	0.5740	0.9293
		8	0.0019	0.0001	0.0004	0.0004	1.1331	1.1331	-0.0028	-0.0000	-0.0005				
		9	-0.0004	-0.0000	-0.0003	-0.0003	3.3556	3.3556	-0.0037	-0.0004	-0.0005				
793	9	7	0.599	3.06	-0.000	0.000	-0.1451	-0.000	-2.91	0.05	-2.94	0.3319	0.5812	0.1855	0.6065
		8	0.0026	-0.0000	0.0003	0.0003	0.5775	0.5775	0.0113	0.0007	0.0013				
		9	0.0007	-0.0000	-0.0003	-0.0003	-2.3786	-2.3786	0.0103	0.0003	0.0012				
793	10	7	0.598	6.18	-0.000	0.000	-0.3707	-0.002	-2.86	0.06	-2.89	0.3037	0.6086	0.2704	0.6174
		8	0.0019	-0.0001	0.0001	0.0001	0.4902	0.4902	0.0361	0.0021	0.0044				
		9	-0.0000	0.0001	-0.0000	-0.0000	1.0283	1.0283	0.0366	0.0019	0.0045				

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key											
795	1	0.598	-3.07	-0.000	-0.2705	-0.9253	0.96	0.05	0.92	0.2559	0.6219	0.3473	0.5978
	8	0.0011	-0.0000	0.0002	-0.1659	0.5169	-0.0170	-0.0008	-0.0021				
	9	-0.0019	0.0000	-0.0002	-0.1659	0.5169	-0.0153	-0.0010	-0.0018				
795	2	0.599	-0.01	-0.000	0.5551	1.8828	1.01	0.05	0.56	0.3953	0.6358	0.1659	0.7092
	8	0.0012	0.0001	0.0004	0.4402	2.9316	0.063	0.0005	0.0008				
	9	-0.0004	-0.0000	-0.0002	0.4402	2.9316	0.060	0.0002	0.0008				
795	3	0.598	1.02	-0.000	0.0765	-0.9079	1.03	0.05	0.98	0.3322	0.6190	0.2317	0.6658
	8	0.0022	0.0000	0.0004	0.1358	1.0346	0.149	0.0009	0.0018				
	9	-0.0011	0.0000	-0.0002	0.1358	1.0346	0.133	0.0006	0.0017				
795	4	0.598	3.08	-0.000	0.4915	-0.4266	1.06	0.05	1.01	0.3068	0.6219	0.2618	0.6249
	8	0.0016	0.0001	0.0004	0.4923	1.4311	0.326	0.0020	0.0040				
	9	-0.0006	0.0000	-0.0001	0.4923	1.4311	0.313	0.0016	0.0039				
795	5	0.598	6.21	-0.000	-0.3633	-0.6946	1.12	0.05	1.06	0.3013	0.5794	0.2753	0.5813
	8	0.0014	0.0001	0.0001	-0.5179	-1.1041	0.560	0.0033	0.0064				
	9	0.0006	0.0000	-0.0001	-0.5179	-1.1041	0.571	0.0031	0.0066				
795	6	0.597	9.30	-0.000	-0.4795	-0.4388	1.15	0.05	1.09	0.2732	0.5457	0.2549	0.5461
	8	0.0020	0.0001	0.0001	-0.4792	-0.2005	0.757	0.0041	0.0082				
	9	-0.0018	0.0001	0.0000	-0.4792	-0.2005	0.759	0.0038	0.0082				
795	7	0.597	12.46	-0.000	-0.6667	-0.7611	1.15	0.05	1.09	0.2002	0.5377	0.2010	0.5303
	8	0.0011	0.0001	0.0001	-0.2803	-0.0541	0.879	0.0035	0.0094				
	9	-0.0031	0.0001	0.0000	-0.2803	-0.0541	0.892	0.0035	0.0094				
795	8	0.598	-0.02	-0.000	0.1026	1.0757	1.01	0.05	0.97	0.3793	0.6090	0.1711	0.7062
	8	0.0019	0.0000	0.0004	0.9096	9.9851	0.065	0.0004	0.0007				
	9	-0.0001	-0.0000	-0.0003	0.9096	9.9851	0.061	0.0002	0.0008				
796	1	0.599	-3.09	-0.000	-0.1290	-0.3517	2.96	0.04	3.02	0.1994	0.6686	0.4590	0.5628
	8	0.0009	-0.0000	0.0002	-0.1291	2.2164	-0.069	-0.0002	-0.0005				
	9	-0.0006	-0.0000	-0.0003	-0.1291	2.2164	-0.050	-0.0004	-0.0005				
796	2	0.599	-0.00	-0.000	0.0909	-0.9619	3.01	0.04	3.07	0.3316	0.6397	0.2450	0.6647
	8	0.0021	0.0000	0.0004	0.2656	0.7117	0.167	0.0011	0.0021				
	9	-0.0011	0.0000	-0.0001	0.2656	0.7117	0.160	0.0007	0.0021				
796	3	0.599	0.90	-0.000	0.1225	-0.9681	3.03	0.05	3.09	0.3116	0.6341	0.2603	0.6409
	8	0.0021	0.0000	0.0004	0.1189	-0.3763	0.258	0.0016	0.0032				
	9	-0.0005	0.0000	-0.0002	0.1189	-0.3763	0.248	0.0012	0.0031				

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																			
796	4	7	0.599	3.10	-0.004	0.1845	-0.9850	3.98	0.04	3.12	0.3038	0.6142	0.0020	0.0000	0.0004	0.1845	-0.9850	3.98	0.04	3.12	0.3038	0.6142
		8	-0.0019	0.0001	-0.0000	-0.3644	0.1015	0.0429	0.0023	0.0053	0.2710	0.6260	0.0019	0.0000	0.0000	-0.3644	0.1015	0.0429	0.0023	0.0053	0.2710	0.6260
796	5	7	0.600	6.19	-0.001	-0.3633	-0.6946	3.12	0.05	3.16	0.2857	0.5723	0.0014	0.0001	0.0001	-0.3633	-0.6946	3.12	0.05	3.16	0.2857	0.5723
		8	0.0003	0.0000	-0.0001	1.1617	-1.4974	0.0660	0.0035	0.0074	0.2686	0.5711	0.0003	0.0000	-0.0001	1.1617	-1.4974	0.0660	0.0035	0.0074	0.2686	0.5711
796	6	7	0.600	9.28	-0.002	-0.3149	-0.9795	3.15	0.05	3.18	0.2483	0.5388	0.0010	0.0000	0.0002	-0.3149	-0.9795	3.15	0.05	3.18	0.2483	0.5388
		8	0.0010	0.0000	0.0002	1.0984	0.9795	0.0809	0.0038	0.0089	0.2343	0.5371	0.0010	0.0000	0.0002	1.0984	0.9795	0.0809	0.0038	0.0089	0.2343	0.5371
796	7	7	0.600	12.43	-0.002	-0.5452	-2.2482	3.13	0.05	3.17	0.1814	0.5367	0.0005	0.0000	0.0002	-0.5452	-2.2482	3.13	0.05	3.17	0.1814	0.5367
		8	0.0011	0.0000	-0.0001	-0.1644	0.5700	0.0880	0.0032	0.0093	0.1834	0.5292	0.0011	0.0000	-0.0001	-0.1644	0.5700	0.0880	0.0032	0.0093	0.1834	0.5292
796	8	7	0.599	-0.03	-0.004	0.0083	-0.9373	3.02	0.05	3.06	0.3178	0.6153	0.0021	0.0000	0.0004	0.0083	-0.9373	3.02	0.05	3.06	0.3178	0.6153
		8	0.0021	0.0000	0.0004	-0.2312	0.3034	0.0173	0.0011	0.0021	0.2375	0.6389	0.0021	0.0000	0.0004	-0.2312	0.3034	0.0173	0.0011	0.0021	0.2375	0.6389
797	1	7	0.598	-3.07	-0.002	-0.1790	-0.9373	6.01	0.05	6.00	0.3923	0.6490	0.0011	0.0000	0.0002	-0.1790	-0.9373	6.01	0.05	6.00	0.3923	0.6490
		8	0.0011	0.0000	0.0002	-0.1847	0.3751	0.0066	0.0003	0.0008	0.2050	0.7364	0.0011	0.0000	0.0002	-0.1847	0.3751	0.0066	0.0003	0.0008	0.2050	0.7364
797	2	7	0.598	-0.01	-0.003	-0.1031	-0.9373	6.06	0.04	6.05	0.3047	0.6376	0.0025	0.0000	0.0003	-0.1031	-0.9373	6.06	0.04	6.05	0.3047	0.6376
		8	0.0025	0.0000	0.0003	-0.2264	0.3440	0.0335	0.0017	0.0042	0.2600	0.6421	0.0025	0.0000	0.0003	-0.2264	0.3440	0.0335	0.0017	0.0042	0.2600	0.6421
797	3	7	0.598	1.05	-0.003	-0.3349	-0.9997	6.08	0.04	6.07	0.3058	0.6249	0.0028	0.0000	0.0003	-0.3349	-0.9997	6.08	0.04	6.07	0.3058	0.6249
		8	0.0028	0.0000	0.0003	-0.2363	0.3346	0.0426	0.0022	0.0053	0.2701	0.6353	0.0028	0.0000	0.0003	-0.2363	0.3346	0.0426	0.0022	0.0053	0.2701	0.6353
797	4	7	0.598	3.12	-0.004	-0.1061	-0.9949	6.11	0.04	6.10	0.2962	0.5836	0.0022	0.0000	0.0004	-0.1061	-0.9949	6.11	0.04	6.10	0.2962	0.5836
		8	0.0022	0.0000	0.0004	-0.4725	0.5747	0.0571	0.0031	0.0067	0.2766	0.5915	0.0022	0.0000	0.0004	-0.4725	0.5747	0.0571	0.0031	0.0067	0.2766	0.5915
797	5	7	0.598	6.21	-0.002	-0.3511	-0.9993	6.14	0.05	6.13	0.2677	0.5504	0.0016	0.0000	0.0002	-0.3511	-0.9993	6.14	0.05	6.13	0.2677	0.5504
		8	0.0016	0.0001	0.0002	-0.8169	0.3941	0.0765	0.0038	0.0084	0.2519	0.5477	0.0016	0.0000	0.0002	-0.8169	0.3941	0.0765	0.0038	0.0084	0.2519	0.5477
797	6	7	0.598	9.31	-0.002	-0.3565	-0.9949	6.14	0.05	6.13	0.2007	0.5425	0.0007	0.0000	0.0002	-0.3565	-0.9949	6.14	0.05	6.13	0.2007	0.5425
		8	0.0007	0.0000	0.0002	-1.1793	1.6493	0.0867	0.0035	0.0094	0.2016	0.5354	0.0007	0.0000	0.0002	-1.1793	1.6493	0.0867	0.0035	0.0094	0.2016	0.5354

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																		
801	7	7	1.251	6.22	-0.002	-0.4383	-0.02	-2.87	0.06	-2.90	0.1232	0.6524	0.0013	0.0001	0.0002	0.8260	0.390	0.0009	0.0050	0.0960	0.6510
			0.0015	0.0002	0.0000	0.8371	0.3004	0.0395	0.0007	0.0051											
A01	8	7	1.250	9.35	-0.002	-0.4462	-0.02	-2.82	0.06	-2.86	0.1156	0.6458	0.0012	0.0001	0.0002	0.8122	0.658	0.0015	0.0085	0.1026	0.6447
			0.0000	0.0001	-0.0000	12.4738	-5.4787	0.0662	0.0013	0.0085											
A01	9	7	1.252	12.60	-0.002	-1.4295	-0.02	-2.79	0.06	-2.83	0.1046	0.6297	0.0001	0.0001	0.0001	4.6217	0.882	0.0018	0.0111	0.0949	0.6270
			0.0003	0.0002	-0.0001	-3.2695	1.9255	0.0888	0.0016	0.0111											
801	10	7	1.250	-0.07	-0.002	-0.3379	-0.02	-2.96	0.06	-2.98	0.1238	0.7216	0.0031	0.0002	0.0004	0.7154	0.143	-0.0003	-0.0020	0.2384	0.7145
			0.0011	0.0002	-0.0001	-1.0988	0.7418	-0.0148	-0.0007	-0.0021											
802	3	7	1.250	-3.14	-0.002	-0.5467	-0.02	-0.10	0.05	-0.07	0.1192	0.6787	0.0016	0.0001	0.0002	0.7444	0.268	-0.0006	-0.0036	0.1750	0.6792
			0.0018	0.0002	-0.0002	-0.5995	0.5959	-0.0261	-0.0009	-0.0035											
A02	4	7	1.250	-0.05	-0.002	-0.2721	-0.02	-0.05	0.04	-0.03	0.4068	0.3882	0.0026	0.0001	0.0004	0.8462	0.010	-0.0002	0.0000	0.4068	0.3882
			0.0005	0.0002	-0.0002	-2.0242	2.1614	0.0006	-0.0002	0.0000											
A02	5	7	1.251	1.01	-0.002	-0.2560	-0.01	-0.04	0.04	-0.02	0.1525	0.6526	0.0032	0.0001	0.0004	0.7050	0.106	0.0003	0.0012	0.0141	0.7004
			0.0009	0.0002	-0.0001	-1.2950	0.8028	0.0086	0.0000	0.0012											
A02	6	7	1.250	3.15	-0.002	-0.0601	-0.01	-0.01	0.05	0.00	0.1199	0.6511	0.0021	0.0001	0.0003	0.8028	0.302	0.0007	0.0039	0.0893	0.6617
			0.0002	0.0002	-0.0000	-4.8793	0.7153	0.0275	0.0004	0.0036											
802	7	7	1.250	6.26	-0.002	-0.5773	-0.02	0.03	0.05	0.04	0.1117	0.6578	0.0011	0.0001	0.0002	0.8184	0.562	0.0012	0.0073	0.0972	0.6548
			0.0011	0.0002	0.0000	1.2829	0.3925	0.0567	0.0011	0.0074											
A02	8	7	1.250	9.41	-0.002	-0.4769	-0.02	0.07	0.04	0.07	0.1063	0.6393	0.0012	0.0001	0.0002	0.6661	0.809	0.0017	0.0103	0.1063	0.6393
			0.0006	0.0002	-0.0000	-1.8800	0.6981	0.0814	0.0015	0.0102											
802	9	7	1.249	12.63	-0.002	-0.9371	-0.02	0.10	0.05	0.10	0.0954	0.6235	0.0004	0.0001	0.0002	1.0600	0.1014	0.0019	0.0126	0.0954	0.6235
			0.0005	0.0002	-0.0001	-2.2077	1.1168	0.1018	0.0019	0.0125											

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																			
802	10	7	1.250	-0.001	-0.004	-0.3223	-0.02	-0.07	0.05	-0.03	0.3348	0.1772	0.0029	0.0002	0.0004	-0.3223	-0.02	-0.07	0.05	-0.03	0.3348	0.1772
		8	0.0029	-0.0002	0.0004	-1.3089	0.7114	0.0009	0.0002	0.0000	-2.4557	0.3107	0.0009	0.0002	0.0004	-1.3089	0.7114	0.0009	0.0002	0.0000	-2.4557	0.3107
		9	-0.0009	0.0002	-0.0002	-1.3089	1.0604	0.0005	-0.0002	0.0000	-2.4557	0.3107	0.0005	-0.0002	0.0000	-1.3089	1.0604	0.0005	-0.0002	0.0000	-2.4557	0.3107
803	1	7	1.250	-3.11	-0.002	-0.5406	-0.02	0.95	0.04	0.92	0.1302	0.6884	0.0018	0.0002	0.0002	-0.5406	-0.02	0.95	0.04	0.92	0.1302	0.6884
		8	0.0018	-0.0002	0.0002	-0.5406	0.6616	-0.0201	-0.0005	-0.0027	0.2005	0.6799	0.0018	0.0002	0.0002	-0.5406	-0.02	0.95	0.04	0.92	0.2005	0.6799
		9	-0.0013	0.0002	-0.0002	-0.7624	1.0938	-0.0196	-0.0007	-0.0026	0.2005	0.6799	-0.0013	0.0002	-0.0002	-0.7624	1.0938	-0.0196	-0.0007	-0.0026	0.2005	0.6799
803	2	7	1.250	-0.04	-0.004	-0.2536	-0.02	1.00	0.05	0.96	0.1749	0.6396	0.0024	0.0004	0.0004	-0.2536	-0.02	1.00	0.05	0.96	0.1749	0.6396
		8	0.0024	-0.0001	0.0004	-0.2536	0.9414	0.0062	0.0002	0.0007	-0.0734	0.6396	0.0024	0.0004	0.0004	-0.2536	-0.02	1.00	0.05	0.96	-0.0734	0.6396
		9	-0.0008	0.0002	-0.0002	-1.4974	1.4998	0.0049	-0.0000	0.0006	-0.0734	0.6396	-0.0008	0.0002	-0.0002	-1.4974	1.4998	0.0049	-0.0000	0.0006	-0.0734	0.6396
803	3	7	1.249	1.02	-0.004	-0.2166	-0.02	1.03	0.05	0.97	0.1333	0.6450	0.0028	0.0004	0.0004	-0.2166	-0.02	1.03	0.05	0.97	0.1333	0.6450
		8	0.0028	-0.0001	0.0004	-0.2166	0.7852	0.0165	0.0004	0.0021	0.0490	0.6450	0.0028	0.0004	0.0004	-0.2166	-0.02	1.03	0.05	0.97	0.1333	0.6450
		9	-0.0005	0.0002	-0.0001	-2.0281	1.6344	0.0146	0.0001	0.0019	0.0490	0.6450	-0.0005	0.0002	-0.0001	-2.0281	1.6344	0.0146	0.0001	0.0019	0.0490	0.6450
803	4	7	1.250	3.15	-0.003	-0.0993	-0.02	1.06	0.05	1.01	0.1175	0.6632	0.0021	0.0003	0.0003	-0.0993	-0.02	1.06	0.05	1.01	0.1175	0.6632
		8	0.0021	-0.0002	0.0003	-0.0993	0.9299	0.0362	0.0008	0.0048	0.0922	0.6632	0.0021	0.0003	0.0003	-0.0993	-0.02	1.06	0.05	1.01	0.1175	0.6632
		9	-0.0003	0.0002	-0.0000	-4.8108	0.1515	0.0338	0.0006	0.0044	0.0922	0.6632	-0.0003	0.0002	-0.0000	-4.8108	0.1515	0.0338	0.0006	0.0044	0.0922	0.6632
803	5	7	1.249	6.29	-0.001	-0.5858	-0.02	1.10	0.05	1.04	0.1099	0.6577	0.0010	0.0001	0.0001	-0.5858	-0.02	1.10	0.05	1.04	0.1099	0.6577
		8	0.0010	-0.0001	0.0001	-0.5858	0.8504	0.0621	0.0013	0.0081	0.0950	0.6577	0.0010	0.0001	0.0001	-0.5858	-0.02	1.10	0.05	1.04	0.1099	0.6577
		9	0.0006	0.0003	0.0001	-2.8529	1.3133	0.0631	0.0011	0.0081	0.0950	0.6577	0.0006	0.0003	0.0001	-2.8529	1.3133	0.0631	0.0011	0.0081	0.0950	0.6577
803	6	7	1.251	9.42	-0.001	-0.5528	-0.02	1.14	0.05	1.07	0.1030	0.6358	0.0011	0.0002	0.0002	-0.5528	-0.02	1.14	0.05	1.07	0.1030	0.6358
		8	0.0011	-0.0001	0.0001	-0.5528	0.7262	0.0859	0.0017	0.0109	0.0954	0.6358	0.0011	0.0002	0.0002	-0.5528	-0.02	1.14	0.05	1.07	0.1030	0.6358
		9	-0.0007	0.0002	-0.0000	-1.6241	0.6268	0.0859	0.0016	0.0108	0.0954	0.6358	-0.0007	0.0002	-0.0000	-1.6241	0.6268	0.0859	0.0016	0.0108	0.0954	0.6358
803	7	7	1.251	12.62	-0.001	-0.7656	-0.02	1.17	0.05	1.10	0.0937	0.6205	0.0008	0.0001	0.0001	-0.7656	-0.02	1.17	0.05	1.10	0.0937	0.6205
		8	0.0008	-0.0001	0.0000	-0.7656	0.3206	0.1054	0.0019	0.0130	0.0867	0.6205	0.0008	0.0001	0.0001	-0.7656	-0.02	1.17	0.05	1.10	0.0937	0.6205
		9	-0.0007	0.0002	-0.0000	-1.7532	0.5884	0.1059	0.0018	0.0130	0.0867	0.6205	-0.0007	0.0002	-0.0000	-1.7532	0.5884	0.1059	0.0018	0.0130	0.0867	0.6205
803	8	7	1.250	-0.04	-0.003	-0.3705	-0.01	1.00	0.05	0.96	0.1630	0.6372	0.0029	0.0002	0.0002	-0.3705	-0.01	1.00	0.05	0.96	0.1630	0.6372
		8	0.0029	-0.0002	0.0003	-0.3705	0.6622	0.0060	0.0002	0.0006	-0.0657	0.6372	0.0029	0.0002	0.0002	-0.3705	-0.01	1.00	0.05	0.96	-0.0657	0.6372
		9	-0.0008	0.0002	-0.0002	-1.5067	1.3828	0.0045	-0.0000	0.0006	-0.0657	0.6372	-0.0008	0.0002	-0.0002	-1.5067	1.3828	0.0045	-0.0000	0.0006	-0.0657	0.6372
804	1	7	1.250	-3.11	-0.001	-0.5467	-0.02	2.94	0.04	3.03	0.1239	0.7142	0.0017	0.0001	0.0002	-0.5467	-0.02	2.94	0.04	3.03	0.1239	0.7142
		8	0.0017	-0.0001	0.0002	-0.5467	0.7644	-0.0086	-0.0002	-0.0012	0.3221	0.7142	0.0017	0.0001	0.0002	-0.5467	-0.02	2.94	0.04	3.03	0.3221	0.7142
		9	-0.0016	0.0002	-0.0002	-0.7092	0.8461	-0.0079	-0.0005	-0.0010	0.3221	0.7142	-0.0016	0.0002	-0.0002	-0.7092	0.8461	-0.0079	-0.0005	-0.0010	0.3221	0.7142
804	2	7	1.250	-0.03	-0.003	-0.3187	-0.02	2.99	0.05	3.08	0.1412	0.6652	0.0027	0.0001	0.0002	-0.3187	-0.02	2.99	0.05	3.08	0.1412	0.6652
		8	0.0027	-0.0001	0.0003	-0.3187	0.7146	0.0174	0.0004	0.0023	0.0704	0.6652	0.0027	0.0001	0.0002	-0.3187	-0.02	2.99	0.05	3.08	0.0704	0.6652
		9	-0.0012	0.0002	-0.0002	-1.1170	0.8369	0.0162	0.0002	0.0021	0.0704	0.6652	-0.0012	0.0002	-0.0002	-1.1170	0.8369	0.0162	0.0002	0.0021	0.0704	0.6652

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																		
804	3	7	1.250	1.05	-0.004	-0.2550	-0.01	3.01	0.05	3.09	0.1187	0.6705	0.0029	-0.0001	0.0004	-0.6835	0.289	0.006	0.0038	0.0890	0.6687
			-0.0005	0.0002	0.0001	-2.0958	1.5786	0.0267	0.0004	0.0035	0.0004	0.0035									
A04	4	7	1.251	3.14	-0.003	-0.1511	-0.02	3.04	0.05	3.12	0.1102	0.6674	0.0024	0.0002	0.0010	0.7019	0.486	0.010	0.0064	0.0953	0.6650
			0.0007	0.0002	-0.0001	1.4357	-0.9304	0.0466	0.0002	0.0008	0.0008	0.0062									
A04	5	7	1.250	6.31	-0.001	-0.4989	-0.02	3.08	0.05	3.15	0.1048	0.6514	0.0009	0.0001	0.0015	1.0084	0.727	0.015	0.0094	0.0942	0.6544
			0.0019	0.0002	0.0000	0.7681	0.1235	0.0740	0.0002	0.0013	0.0013	0.0095									
A04	6	7	1.251	9.44	-0.001	-0.5390	-0.02	3.12	0.05	3.18	0.0991	0.6329	0.0012	0.0001	0.0018	0.6598	0.950	0.018	0.0120	0.0910	0.6271
			0.0001	0.0002	-0.0001	-6.7692	3.6987	0.0956	0.0001	0.0017	0.0017	0.0119									
A04	7	7	1.250	12.64	-0.001	-0.7556	-0.02	3.14	0.05	3.20	0.0924	0.6136	0.0007	0.0001	0.0020	0.413	1.128	0.020	0.0138	0.0848	0.6063
			0.0005	0.0002	-0.0001	-2.2391	1.1248	0.1137	0.0001	0.0019	0.0019	0.0138									
A04	8	7	1.250	0.00	-0.002	-0.3039	-0.02	2.99	0.05	3.08	0.1343	0.6567	0.0026	0.0001	0.0004	0.7629	0.176	0.004	0.0023	0.0725	0.6705
			0.0012	0.0002	-0.0002	-1.1771	0.8132	0.0163	0.0002	0.0002	0.0002	0.0021									
A05	1	7	1.249	-3.08	-0.002	-0.5860	-0.02	6.01	0.05	5.99	0.2127	0.7467	0.0014	0.0001	0.0002	0.9131	0.064	0.002	0.0009	0.0265	0.7441
			0.0014	0.0002	-0.0003	-0.7679	1.0406	0.0061	0.0002	0.0000	0.0000	0.0009									
A05	2	7	1.249	0.00	-0.003	-0.3475	-0.02	6.06	0.05	6.04	0.1135	0.6825	0.0028	0.0001	0.0008	0.6033	0.369	0.008	0.0050	0.0890	0.6804
			0.0010	0.0002	-0.0002	-1.2769	1.2060	0.0351	0.0002	0.0006	0.0006	0.0047									
A05	3	7	1.249	1.08	-0.004	-0.2165	-0.01	6.07	0.05	6.05	0.1091	0.6773	0.0027	0.0001	0.0010	0.7405	0.474	0.010	0.0064	0.0944	0.6724
			0.0003	0.0002	-0.0002	-3.9887	3.4029	0.0452	0.0002	0.0008	0.0008	0.0061									
A05	4	7	1.250	3.20	-0.003	-0.1201	-0.02	6.11	0.05	6.08	0.1053	0.6623	0.0024	0.0001	0.0013	0.7227	0.653	0.013	0.0086	0.0948	0.6590
			0.0001	0.0003	-0.0000	-11.5264	2.9623	0.0634	0.0003	0.0012	0.0012	0.0083									
A05	5	7	1.250	6.34	-0.001	-0.5556	-0.02	6.14	0.05	6.11	0.0984	0.6419	0.0009	0.0001	0.0017	0.9229	0.876	0.017	0.0112	0.0899	0.6338
			0.0007	0.0003	0.0000	-2.1176	0.4346	0.0885	0.0001	0.0015	0.0015	0.0112									

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																								
A05	6	7	8	9	10	11	12	13	14	15	16	17	18	19	20	21	22	23	24	25	26	27	28	29	30	
A05	7	8	9	10	11	12	13	14	15	16	17	18	19	20	21	22	23	24	25	26	27	28	29	30	31	32
A05	8	9	10	11	12	13	14	15	16	17	18	19	20	21	22	23	24	25	26	27	28	29	30	31	32	33
A06	1	2	3	4	5	6	7	8	9	10	11	12	13	14	15	16	17	18	19	20	21	22	23	24	25	26
A06	2	3	4	5	6	7	8	9	10	11	12	13	14	15	16	17	18	19	20	21	22	23	24	25	26	27
A06	3	4	5	6	7	8	9	10	11	12	13	14	15	16	17	18	19	20	21	22	23	24	25	26	27	28
A06	4	5	6	7	8	9	10	11	12	13	14	15	16	17	18	19	20	21	22	23	24	25	26	27	28	29
A06	5	6	7	8	9	10	11	12	13	14	15	16	17	18	19	20	21	22	23	24	25	26	27	28	29	30
A06	6	7	8	9	10	11	12	13	14	15	16	17	18	19	20	21	22	23	24	25	26	27	28	29	30	31
A06	7	8	9	10	11	12	13	14	15	16	17	18	19	20	21	22	23	24	25	26	27	28	29	30	31	32
A06	8	9	10	11	12	13	14	15	16	17	18	19	20	21	22	23	24	25	26	27	28	29	30	31	32	33

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key														
807	1	7	1.250	-2.9A	-0.002	-0.6773	-0.02	15.10	0.04	15.07	0.1023	0.6035					
		8	0.0011	-0.0001	0.0002	-0.6773	0.9971	0.0594	0.0012	0.0081	0.0920	0.6793					
		9	-0.0016	0.0002	-0.0003	-0.6778	0.9854	0.0594	0.0010	0.0080							
807	2	7	1.249	0.09	-0.00	-0.2468	-0.02	15.14	0.05	15.10	0.0979	0.6512					
		8	0.0023	-0.0001	0.0003	-0.2468	0.8247	0.0846	0.0016	0.0110	0.0905	0.6522					
		9	-0.0005	0.0002	-0.0002	-2.0810	2.4317	0.0842	0.0015	0.0109							
807	3	7	1.250	1.13	-0.00	-0.0768	-0.01	15.15	0.05	15.11	0.0941	0.6398					
		8	0.0024	-0.0000	0.0004	-0.0768	0.8189	0.0926	0.0017	0.0118	0.0878	0.6405					
		9	-0.0006	0.0002	-0.0002	-1.7875	1.7214	0.0918	0.0016	0.0117							
807	4	7	1.251	3.24	-0.00	0.1707	-0.01	15.17	0.05	15.12	0.0922	0.6258					
		8	0.0015	0.0000	0.0003	0.1707	1.0414	0.1054	0.0019	0.0131	0.0868	0.6255					
		9	-0.0001	0.0003	-0.0001	-9.8438	5.6670	0.1050	0.0018	0.0130							
807	5	7	1.250	6.35	-0.00	-0.4239	-0.02	15.19	0.05	15.15	0.0974	0.6000					
		8	0.0012	-0.0001	0.0001	-0.4239	0.6652	0.1165	0.0022	0.0139	0.0986	0.5903					
		9	0.0011	0.0003	0.0000	1.4761	0.1433	0.1178	0.0023	0.0139							
807	6	7	1.251	9.48	-0.00	-0.4406	-0.02	15.21	0.05	15.16	0.0880	0.5916					
		8	0.0013	-0.0001	0.0001	-0.4406	0.4924	0.1284	0.0022	0.0152	0.0868	0.5852					
		9	-0.0002	0.0002	-0.0001	-4.9507	3.6479	0.1300	0.0022	0.0152							
807	10	7	1.249	12.64	0.00	-0.6742	-0.02	15.21	0.05	15.16	0.0808	0.5742					
		8	0.0005	-0.0000	0.0001	-0.6742	1.2483	0.1350	0.0021	0.0158	0.0808	0.5883					
		9	-0.0006	0.0001	-0.0001	-1.4712	0.9910	0.1358	0.0021	0.0156	0.0783	0.5742					
807	11	7	1.252	0.07	-0.00	-0.2810	-0.02	15.14	0.06	15.10	0.1015	0.6518					
		8	0.0025	-0.0001	0.0004	-0.2810	0.7928	0.0840	0.0017	0.0109	0.1015	0.6518					
		9	-0.0015	0.0002	-0.0001	-0.8901	0.3842	0.0843	0.0015	0.0109	0.0902	0.6497					
80A	1	7	0.898	-3.02	-0.00	-0.4587	-0.02	15.10	0.06	15.08	0.2272	0.6131					
		8	0.0025	-0.0002	0.0003	-0.4587	0.7646	0.0598	0.0027	0.0073	0.2272	0.6135					
		9	-0.0020	0.0002	-0.0002	-0.5148	0.5467	0.0615	0.0025	0.0075	0.2072	0.6135					
80A	2	7	0.901	0.03	-0.00	-0.2024	-0.02	15.12	0.05	15.10	0.2035	0.5777					
		8	0.0023	-0.0000	0.0003	-0.2024	0.8007	0.0795	0.0032	0.0091	0.2035	0.5777					
		9	-0.0013	0.0001	-0.0002	-0.6666	0.7540	0.0803	0.0030	0.0091	0.1894	0.5718					
80A	3	7	0.901	1.07	-0.00	-0.1443	-0.02	15.13	0.06	15.11	0.1837	0.5731					
		8	0.0022	-0.0000	0.0002	-0.1443	0.6387	0.0855	0.0031	0.0098	0.1837	0.5731					
		9	-0.0015	0.0001	-0.0001	-0.6186	0.5255	0.0856	0.0030	0.0097	0.1803	0.5667					

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd																			
A08	4	7	0.699	3.15	-0.000	-0.000	-0.1812	-0.02	15.13	0.06	15.11	0.1618	0.5655							
		8	0.0020	-0.0001	0.0002	-0.0002	0.5671	0.0941	0.030	0.0106	0.1572	0.5592								
		9	-0.0005	0.0001	-0.0002	-1.4688	1.7663	0.0956	0.0030	0.0106										
A0A	5	7	0.901	6.24	-0.000	-1.1200	-0.02	15.13	0.06	15.10	0.1245	0.5698								
		8	0.0006	-0.0001	0.0002	-73.3523	1.7678	0.1013	0.025	0.0115	0.1213	0.5566								
		9	-0.0009	0.0002	-0.0000	11.3823	1.1041	0.1041	0.0025	0.0115										
A0P	6	7	0.900	9.32	-0.000	-1.4641	-0.02	15.14	0.05	15.11	0.1158	0.5661								
		8	0.0007	-0.0002	0.0001	-0.8001	0.9172	0.1109	0.025	0.0124	0.1164	0.5518								
		9	-0.0012	0.0002	-0.0000	0.1770	0.1132	0.1132	0.0026	0.0124										
A0R	7	7	0.898	12.48	0.000	-1.6365	-0.03	15.14	0.05	15.11	0.0984	0.5701								
		8	0.0005	-0.0001	0.0001	-1.3886	1.2302	0.1178	0.023	0.0134	0.0936	0.5613								
		9	-0.0020	0.0001	-0.0001	0.3373	0.1193	0.1193	0.0022	0.0133										
A0S	8	7	0.899	0.02	-0.000	-0.2445	-0.02	15.12	0.05	15.10	0.2028	0.5757								
		8	0.0026	-0.0001	0.0003	-0.6319	0.6713	0.0796	0.032	0.0091	0.1897	0.5704								
		9	-0.0018	0.0002	-0.0001	0.3397	0.3397	0.0803	0.0030	0.0091										
A0T	1	7	0.899	3.04	-0.000	-0.3728	-0.02	9.09	0.05	9.01	0.2662	0.6829								
		8	0.0021	-0.0001	0.0003	-0.3952	0.9010	0.268	0.014	0.0036	0.2119	0.6770								
		9	-0.0021	0.0001	-0.0002	0.6517	0.6517	0.0272	0.0011	0.0036										
A0U	2	7	0.899	0.00	-0.000	-0.2361	-0.02	9.14	0.06	9.05	0.2321	0.6172								
		8	0.0030	-0.0001	0.0005	-0.6465	0.8551	0.552	0.025	0.0068	0.2023	0.6215								
		9	-0.0014	0.0001	-0.0001	0.5142	0.5142	0.0550	0.0022	0.0068										
A0V	3	7	0.900	1.06	-0.000	-0.2489	-0.02	9.15	0.05	9.06	0.2261	0.6056								
		8	0.0031	-0.0001	0.0004	-0.5743	0.6954	0.623	0.028	0.0075	0.2035	0.6027								
		9	-0.0016	0.0001	-0.0001	0.4337	0.4337	0.0620	0.0025	0.0074										
A0W	4	7	0.900	3.13	-0.000	-0.0115	-0.02	9.17	0.05	9.08	0.2120	0.5761								
		8	0.0015	0.0000	0.0002	-0.7530	0.7739	0.757	0.032	0.0087	0.1980	0.5773								
		9	-0.0012	0.0001	-0.0001	0.4046	0.4046	0.0760	0.0030	0.0087										
A0X	5	7	0.899	6.21	-0.000	-0.7825	-0.02	9.18	0.05	9.08	0.1712	0.5659								
		8	0.0004	-0.0000	0.0001	19.9981	1.8969	0.908	0.031	0.0102	0.1655	0.5565								
		9	0.0000	0.0002	-0.0000	-2.5969	0.0931	0.0931	0.0030	0.0103										
A0Y	6	7	0.898	9.30	-0.000	-2.4148	-0.02	9.18	0.06	9.09	0.1280	0.5689								
		8	0.0004	-0.0001	0.0001	-0.7253	1.8760	0.1007	0.025	0.0114	0.1255	0.5553								
		9	-0.0013	0.0002	-0.0000	0.1580	0.1580	0.1042	0.0026	0.0115										

See Key

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key											
809	7	0.899	12.47	0.00	2.8319	-0.02	9.19	0.06	9.125	0.1155	0.5657		
	8	-0.0002	-0.0001	0.0001	-0.4074	-2.4485	0.1107	0.0025	0.0125	0.1177	0.5490		
	9	-0.0018	0.0001	-0.0001	0.3853	0.1144	0.1144	0.0026	0.0125	0.1177	0.5490		
A09	7	0.899	0.03	-0.00	-0.1725	-0.02	9.14	0.05	9.04	0.2313	0.6148		
	8	0.0028	-0.0000	0.0005	-0.6798	0.9865	0.0551	0.0025	0.0067	0.2042	0.6225		
	9	-0.0014	0.0001	-0.0001	0.5911	0.5911	0.0547	0.0022	0.0068	0.2042	0.6225		
A10	7	0.898	-3.05	-0.00	-0.3222	-0.02	6.02	0.05	6.02	0.3732	0.7105		
	8	0.0021	-0.0001	0.0003	-0.4211	0.9205	0.0082	0.0006	0.0011	0.2496	0.7259		
	9	-0.0025	0.0002	-0.0002	0.4768	0.4768	0.0075	0.0003	0.0011	0.2496	0.7259		
A10	7	0.899	0.01	-0.00	-0.1708	-0.02	6.07	0.05	6.06	0.2388	0.6553		
	8	0.0030	-0.0001	0.0005	-0.6709	0.9529	0.0397	0.0018	0.0052	0.2079	0.6585		
	9	-0.0013	0.0001	-0.0001	0.5920	0.5920	0.0385	0.0016	0.0050	0.2079	0.6585		
A10	7	0.899	1.04	-0.00	-0.1506	-0.02	6.09	0.05	6.06	0.2349	0.6306		
	8	0.0033	-0.0001	0.0005	-0.6618	0.8245	0.0485	0.0022	0.0061	0.2050	0.6455		
	9	-0.0012	0.0001	-0.0001	0.6171	0.6171	0.0474	0.0019	0.0061	0.2050	0.6455		
A10	7	0.898	3.13	-0.00	-0.0371	-0.02	6.11	0.05	6.08	0.2214	0.5970		
	8	0.0016	-0.0000	0.0002	-0.7892	0.8087	0.0642	0.0028	0.0076	0.2052	0.5979		
	9	-0.0012	0.0001	-0.0001	0.5358	0.5358	0.0635	0.0026	0.0076	0.2052	0.5979		
A10	7	0.901	6.24	-0.00	-3.1345	-0.02	6.13	0.05	6.10	0.1899	0.5737		
	8	0.0000	-0.0000	0.0000	29.6982	17.7876	0.0837	0.0031	0.0096	0.1863	0.5662		
	9	0.0000	0.0000	-0.0000	-7.2004	-7.2004	0.0842	0.0031	0.0096	0.1863	0.5662		
A10	7	0.898	9.30	-0.00	-3.3072	-0.03	6.14	0.06	6.11	0.1488	0.5657		
	8	0.0001	-0.0001	0.0002	5.4821	5.4821	0.0969	0.0028	0.0109	0.1453	0.5547		
	9	-0.0008	0.0001	-0.0001	0.7275	0.7275	0.0998	0.0029	0.0110	0.1453	0.5547		
A10	7	0.900	12.46	0.00	1.9418	-0.03	6.14	0.05	6.12	0.1161	0.5662		
	8	0.0004	-0.0001	0.0000	-0.3768	-0.3509	0.1061	0.0024	0.0120	0.1199	0.5535		
	9	-0.0021	0.0001	-0.0001	0.7304	0.7304	0.1100	0.0026	0.0121	0.1199	0.5535		
A10	7	0.899	-0.01	-0.00	-0.2103	-0.02	6.07	0.05	6.05	0.2428	0.6543		
	8	0.0034	-0.0001	0.0005	-0.7004	0.7735	0.0393	0.0019	0.0051	0.2125	0.6573		
	9	-0.0012	0.0001	-0.0001	0.7300	0.7300	0.0379	0.0016	0.0049	0.2125	0.6573		
A11	7	0.898	-3.08	-0.00	-0.2576	-0.03	2.95	0.06	3.03	0.1668	0.7342		
	8	0.0020	-0.0001	0.0003	-0.4359	0.9070	-0.0078	-0.0002	-0.0011	0.3165	0.7306		
	9	-0.0027	0.0002	-0.0002	0.3981	0.3981	-0.0069	-0.0004	-0.0010	0.3165	0.7306		

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																	
A11	2	0.899	-0.000	-0.000	-0.1364	-0.02	2.99	0.05	3.07	0.2635	0.6606	0.0032	0.8533	0.4964	0.0194	0.0010	0.0025	0.1810	0.6550
	8	-0.0015	0.0001	-0.0001	-0.6553	0.8533	0.0192	0.0006	0.0025	0.1810	0.6550								
A11	3	0.900	1.01	-0.000	-0.2027	-0.02	3.01	0.05	3.08	0.2442	0.6537	0.003A	0.7078	0.0301	0.0014	0.0039	0.1985	0.6499	
	6	0.003A	-0.0001	0.0005	-0.5805	0.5338	0.0284	0.0011	0.0037	0.1985	0.6499								
	9	-0.0014	0.0001	-0.0001	-0.6099	0.1681	0.0490	0.0019	0.0061	0.2310	0.6318								
A11	4	0.896	3.11	-0.000	-0.1747	-0.02	3.04	0.05	3.11	0.2310	0.6318	0.0025	0.8096	0.0498	0.0023	0.0062	0.2017	0.6232	
	8	0.0025	-0.0000	0.0004	-0.6099	0.1681	0.0490	0.0019	0.0061	0.2310	0.6318								
	9	-0.0018	0.0002	-0.0000	-0.2466	-0.02	3.08	0.06	3.15	0.2121	0.5848								
A11	5	0.899	6.20	-0.000	-0.4628	-0.0157	3.08	0.06	3.15	0.2121	0.5848	0.000A	0.8096	0.0713	0.0030	0.0083	0.1975	0.5816	
	8	0.000A	0.0000	0.0000	-0.2466	-0.0157	3.08	0.06	3.15	0.2121	0.5848								
	9	-0.0002	0.0002	-0.0000	-0.4628	1.2903	0.0733	0.0028	0.0085	0.1975	0.5816								
A11	6	0.897	9.28	-0.000	-1.9418	-0.03	3.09	0.05	3.16	0.1799	0.5639	0.0005	0.9804	0.0894	0.0032	0.0100	0.1708	0.5568	
	8	0.0002	-0.0000	0.0001	-1.1310	-0.03	3.09	0.05	3.16	0.1799	0.5639								
	9	-0.0005	0.0001	-0.0001	-1.1310	1.4673	0.0918	0.0031	0.0102	0.1708	0.5568								
A11	7	0.897	12.45	0.000	-5.9862	-0.03	3.10	0.05	3.17	0.1299	0.5559	0.0012	4.4648	0.1020	0.0026	0.0115	0.1289	0.5559	
	8	0.0001	-0.0000	0.0001	-0.3849	-0.03	3.10	0.05	3.17	0.1299	0.5559								
	9	-0.0012	0.0000	-0.0002	-0.3849	0.9326	0.1060	0.0027	0.0117	0.1289	0.5559								
A11	8	0.899	-0.02	-0.000	-0.0375	-0.02	3.00	0.05	3.08	0.2641	0.6546	0.0024	1.2260	0.0194	0.0010	0.0025	0.2016	0.6856	
	8	0.0024	-0.0000	0.0005	-0.8768	1.5335	0.0192	0.0007	0.0025	0.2641	0.6546								
	9	-0.0008	0.0001	-0.0002	-0.8768	1.5335	0.0192	0.0007	0.0025	0.2641	0.6546								
A12	1	0.898	-3.09	-0.000	-0.3465	-0.02	0.97	0.05	0.91	0.2016	0.6730	0.0026	0.7895	-0.0205	-0.0008	-0.0028	0.2860	0.6856	
	6	0.0026	-0.0002	0.0002	-0.4491	0.4134	-0.0192	-0.0010	-0.0025	0.2860	0.6730								
	9	-0.0026	0.0002	-0.0002	-0.4491	0.4134	-0.0192	-0.0010	-0.0025	0.2860	0.6730								
A12	2	0.898	-0.03	-0.000	-0.2158	-0.02	1.03	0.05	0.95	0.4218	0.6170	0.0036	0.6588	0.0063	0.0005	0.0007	0.1408	0.6170	
	6	0.0036	-0.0001	0.0004	-0.5779	0.4914	0.0055	0.0001	0.0007	0.1408	0.6170								
	9	-0.0016	0.0001	-0.0001	-0.5779	0.4914	0.0055	0.0001	0.0007	0.1408	0.6170								
A12	3	0.898	1.03	-0.000	-0.0909	-0.02	1.05	0.05	0.96	0.2578	0.6560	0.0034	0.8472	0.0172	0.0008	0.0022	0.1741	0.6560	
	6	0.0034	-0.0000	0.0005	-0.6998	0.7065	0.0150	0.0005	0.0019	0.1741	0.6560								
	9	-0.0012	0.0001	-0.0001	-0.6998	0.7065	0.0150	0.0005	0.0019	0.1741	0.6560								
A12	4	0.897	3.09	-0.000	-0.1477	-0.02	1.08	0.05	1.00	0.2332	0.6409	0.0024	0.9723	0.0382	0.0017	0.0049	0.1981	0.6409	
	6	0.0024	-0.0000	0.0004	-0.7276	0.9723	0.0359	0.0014	0.0045	0.1981	0.6409								
	9	-0.0014	0.0002	-0.0001	-0.7276	0.9723	0.0359	0.0014	0.0045	0.1981	0.6409								

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key											
A12	5	7	0.895	6.17	-0.000	-2.2755	-0.033	1.133	0.067	1.03	0.2182	0.5940	
		8	0.0002	-0.0001	-0.0000	-1.8637	-0.0197	0.0642	0.0025	0.0076	0.1999	0.5926	
		9	-0.0006	0.0002	-0.0000								
A12	6	7	0.898	9.27	-0.000	-6.5273	-0.033	1.13	0.05	1.06	0.1853	0.5729	
		8	0.0000	-0.0000	0.0001	279.5614	0.0847	0.0854	0.0031	0.0096	0.1818	0.5656	
		9	-0.0000	0.0000	-0.0002	24.5890	0.0854		0.0031				
A12	7	7	0.900	12.47	0.000	-70.0807	-0.033	1.14	0.05	1.06	0.1445	0.5643	
		8	0.0000	-0.0001	0.0001	87.8992	0.0996	0.1019	0.0028	0.0112	0.1426	0.5563	
		9	-0.0017	0.0001	-0.0002	0.5643	0.1019		0.0029	0.0113			
A12	8	7	0.901	-0.04	-0.000	-0.02	1.03	0.05	0.05	0.95	0.4054	0.5772	
		8	0.0030	-0.0001	0.0005	0.8582	0.0063	0.0056	0.0005	0.0007	0.1474	0.6627	
		9	-0.0015	0.0001	-0.0001	0.6263	0.0056		0.0001	0.0007			
A13	1	7	0.903	-3.07	-0.000	-0.02	-0.10	0.05	0.05	-0.07	0.2154	0.6637	
		8	0.0020	-0.0001	0.0003	0.8423	-0.0268	-0.011	-0.0011	-0.0035	0.2710	0.6654	
		9	-0.0023	0.0002	-0.0002	0.5466	-0.0251		-0.0013	-0.0033			
A13	2	7	0.899	-0.05	-0.000	-0.02	-0.06	0.06	0.06	-0.02	0.5984	0.2316	
		8	0.0036	-0.0001	0.0004	0.6717	0.0016	0.0002	0.0002	0.0000	0.7313	0.0005	
		9	-0.0014	0.0001	-0.0002	0.7056	0.0005		-0.0000	0.0000			
A13	3	7	0.901	1.02	-0.000	-0.02	-0.04	0.05	0.05	-0.01	0.2952	0.6565	
		8	0.0035	-0.0000	0.0005	0.7759	0.0109	0.0006	0.0006	0.0014	0.1447	0.6220	
		9	-0.0014	0.0001	-0.0001	0.6289	0.0091	0.0002	0.0002	0.0011	0.2321	0.6353	
A13	4	7	0.901	3.12	-0.000	-0.02	-0.01	0.05	0.05	0.01	0.2321	0.6429	
		8	0.0031	-0.0001	0.0004	0.748	0.0316	0.0014	0.0014	0.0040	0.1672	0.6353	
		9	-0.0009	0.0001	-0.0002	1.0811	0.0295	0.0011	0.0011	0.0037	0.2203	0.6037	
A13	5	7	0.902	6.19	-0.000	-0.03	0.03	0.06	0.06	0.05	0.1925	0.6063	
		8	-0.0010	0.0000	-0.0000	0.787	0.0580	0.0025	0.0025	0.0070	0.2203	0.6037	
		9	-0.0002	0.0002	-0.0000	0.7102	0.0601	0.0023	0.0023	0.0072	0.1925	0.6063	
A13	6	7	0.899	9.31	-0.000	-0.03	0.05	0.06	0.06	0.08	0.1944	0.5755	
		8	0.0002	-0.0000	0.0000	0.6109	0.0802	0.0031	0.0031	0.0092	0.1647	0.5711	
		9	-0.0012	0.0001	-0.0000	0.1514	0.0819	0.0030	0.0030	0.0093	0.1647	0.5711	
A13	7	7	0.899	12.43	0.000	-0.03	0.06	0.05	0.05	0.08	0.1571	0.5615	
		8	0.0006	-0.0001	0.0001	1.3827	0.0971	0.0030	0.0030	0.0109	0.1502	0.5615	
		9	-0.0019	0.0001	-0.0001	0.4208	0.0990	0.0029	0.0029	0.0109	0.1502	0.5615	

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																				
813	8 7 6 5	0.901 0.0028 -0.0009	-0.003 -0.0000 0.0001	-0.00 0.0005 -0.0002	-0.1269 -0.8731	-0.02 0.9244 1.4514	-0.05 0.0017 0.0006	0.06 0.0001 -0.0000	-0.03 0.0000 -0.0000	0.5177 0.4958 -0.0281	0.2132 0.0281	814	1 7 6 5	0.899 0.0021 -0.0027	-3.13 -0.0001 0.0002	-0.00 0.0002 -0.0001	-0.3509 -0.4203	-0.03 0.6677 0.2551	-3.02 -0.0461 -0.0426	0.06 0.0020 -0.0021	-3.02 -0.0060 -0.0054	0.2180 0.2576 0.6553 0.6563
814	2 7 6 5	0.900 0.0034 -0.0013	-0.006 -0.0001 0.0001	-0.00 0.0004 -0.0002	-0.1958 -0.7450	-0.02 0.7104 0.7862	-2.96 -0.0149 -0.0145	0.06 0.0005 -0.0009	-2.96 -0.0021 -0.0020	0.1892 0.3225 0.7168 0.7073	814	3 7 6 5	0.900 0.0029 -0.0010	0.98 0.0000 0.0001	-0.00 0.0005 -0.0002	-0.0241 -0.7626	-0.02 0.9810 1.2968	-2.94 -0.0039 -0.0060	0.06 0.0002 -0.0005	-2.96 -0.0007 -0.0010	0.2678 0.4228 0.9404 0.8382	
814	4 7 6 5	0.900 0.0034 -0.0017	3.10 -0.0001 0.0002	-0.00 0.0004 -0.0001	-0.1945 -0.5979	-0.02 0.6866 0.2897	-2.92 0.0133 0.0105	0.06 0.0006 0.0002	-2.94 0.0016 0.0013	0.2404 0.1327 0.6303 0.6262	814	5 7 6 5	0.902 0.0003 -0.0007	6.17 -0.0000 0.0002	-0.00 0.0000 0.0000	-1.1060 -1.6690	-0.03 0.3008 -0.0500	-2.87 0.0427 0.0434	0.06 0.0018 0.0015	-2.89 0.0054 0.0054	0.2154 0.1830 0.6325 0.6274	
814	6 7 6 5	0.899 0.0001 -0.0019	9.28 -0.0000 0.0002	-0.00 0.0000 -0.0000	1.0198 -0.6027	-0.02 2.5333 -0.0859	-2.83 0.0685 0.0698	0.06 0.0028 0.0026	-2.86 0.0080 0.0081	0.2071 0.1883 0.5875 0.5856	814	7 7 6 5	0.901 0.0005 -0.0014	12.45 -0.0001 0.0001	0.00 0.0001 -0.0002	-1.2702 -0.3576	-0.03 1.3432 0.7103	-2.80 0.0894 0.0908	0.06 0.0030 0.0030	-2.84 0.0101 0.0102	0.1730 0.1658 0.5699 0.5623	
814	8 7 6 5	0.902 0.0030 -0.001A	-0.007 -0.0000 0.0002	-0.00 0.0005 -0.0001	-0.1187 -0.6062	-0.02 0.8307 0.4513	-2.96 -0.0150 -0.0146	0.06 0.0005 -0.0009	-2.97 -0.0021 -0.0020	0.1909 0.3247 0.7151 0.7107	815	7 7 6 5	0.600 0.0003 -0.000A	3.10 0.0000 -0.0000	-0.00 0.0002 -0.0003	0.6107 0.2070	-0.02 4.2052 1.7807	-3.01 -0.0394 -0.0372	0.06 0.0021 -0.0023	-3.03 -0.0048 -0.0045	0.2697 0.3120 0.6159 0.6077	
815	8 7 6 5	0.600 0.0019 -0.0009	-0.005 -0.0000 -0.0000	-0.00 0.0004 -0.0002	0.0528 0.0638	-0.02 1.0489 1.5036	-2.96 -0.0112 -0.0116	0.06 0.0005 -0.0008	-2.98 -0.0015 -0.0014	0.2430 0.3698 0.6763 0.6262												

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																	
815	9	0.600	0.0029	-0.0000	0.98	-0.0000	-0.0003	-0.0000	-0.0003	-0.0913	0.6215	-0.0033	0.06	-2.94	-0.0004	0.0005	-0.0005	0.1665	0.8776
		-0.0002	-0.0000	-0.0000	3.07	0.0000	0.0003	0.0000	0.0003	0.0615	-0.02	-0.0039	0.0007	-0.0039	0.0003	0.0012	0.0005	0.5153	0.6674
815	10	0.600	0.0023	0.0000	0.0000	0.0000	-0.0001	-0.0001	-0.0001	-0.3643	0.8201	0.0110	0.06	-2.91	0.0003	0.0013	0.0012	0.3404	0.5968
		-0.0014	0.0001	0.0001	6.18	0.0000	0.0000	0.0000	0.0000	0.0000	0.3621	0.0099	0.0022	0.0099	0.0020	0.0045	0.0012	0.1962	0.6300
815	11	0.600	0.0019	-0.0001	0.0000	0.0000	0.0000	0.0000	0.0000	-0.3707	-0.02	-2.85	0.06	-2.85	0.0022	0.0044	-2.89	0.3031	0.6064
		-0.0009	0.0001	0.0001	9.27	0.0000	0.0000	0.0000	0.0000	-0.7987	-0.0454	0.0366	0.0020	0.0366	0.0020	0.0045	0.0044	0.2734	0.6240
815	12	0.600	0.0006	-0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	-0.6077	-0.02	-2.80	0.06	-2.80	0.0033	0.0070	-2.84	0.2980	0.5689
		-0.0020	0.0001	0.0001	12.44	0.0000	0.0000	0.0000	0.0000	-0.4129	1.7081	0.0599	0.0033	0.0599	0.0033	0.0070	0.0068	0.2770	0.5723
815	13	0.600	0.0007	-0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	-0.9761	-0.02	-2.77	0.06	-2.77	0.0041	0.0085	-2.82	0.2584	0.5354
		-0.0012	0.0000	0.0000	3.09	0.0000	0.0000	0.0000	0.0000	0.0137	1.0491	0.0804	0.0039	0.0804	0.0039	0.0086	0.0085	0.2455	0.5394
815	14	0.600	0.0016	-0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.1254	-0.02	-2.96	0.06	-2.96	0.0005	0.0014	-2.96	0.2475	0.6887
		-0.0005	0.0000	0.0000	6.19	0.0000	0.0000	0.0000	0.0000	0.1131	2.7812	-0.0117	0.0008	-0.0117	0.0008	0.0014	-0.0014	0.3647	0.6345
816	1	0.600	0.0006	-0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.1815	-0.02	-0.11	0.05	-0.11	0.0012	0.0028	-0.07	0.2741	0.6273
		0.0007	0.0000	0.0000	12.44	0.0000	0.0000	0.0000	0.0000	-0.6388	2.9827	-0.0213	0.0014	-0.0213	0.0014	0.0026	-0.0026	0.3307	0.6113
816	2	0.600	0.0019	-0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0040	-0.02	-0.05	0.06	-0.05	0.0002	0.0002	-0.02	0.6262	0.4034
		-0.0010	0.0000	0.0000	3.08	0.0000	0.0000	0.0000	0.0000	-0.0550	1.4174	0.0014	0.0000	0.0014	0.0000	0.0002	0.0002	-0.1629	0.8160
816	3	0.600	0.0027	-0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	-0.0400	-0.02	-0.04	0.06	-0.04	0.0006	0.0011	-0.01	0.3495	0.6190
		-0.0011	0.0000	0.0000	6.19	0.0000	0.0000	0.0000	0.0000	-0.2639	0.7301	0.0095	0.0006	0.0095	0.0006	0.0011	0.0011	0.2100	0.6915
816	4	0.601	0.0013	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.3552	1.4597	0.0260	0.0016	0.0260	0.0013	0.0032	0.01	0.3139	0.6239
		-0.0006	0.0000	0.0000	12.44	0.0000	0.0000	0.0000	0.0000	-0.2892	1.7919	0.0247	0.0013	0.0247	0.0013	0.0031	0.0031	0.2632	0.6317
816	5	0.600	0.0025	-0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	-0.4193	-0.02	-0.04	0.07	-0.04	0.0029	0.0060	0.05	0.3034	0.5902
		-0.0014	0.0000	0.0000	6.19	0.0000	0.0000	0.0000	0.0000	-0.8568	-0.4349	0.0514	0.0029	0.0514	0.0029	0.0063	0.0063	0.2781	0.6018

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																			
816	6	7	0.600	9.27	-0.000	-0.02	0.08	0.06	0.10	0.2793	0.5521	0.0017	-0.0002	0.0000	-0.7461	0.2378	0.0724	0.0040	0.0080	0.2642	0.5497
		8	-0.0029	0.0002	0.0001	-0.1804	0.0725	0.0038	0.0079												
816	7	6	0.600	12.45	-0.001	-0.02	0.08	0.06	0.10	0.2131	0.5357	0.0009	-0.0001	0.0001	-0.7658	0.9605	0.0857	0.0036	0.0091	0.2155	0.5355
		9	-0.0022	0.0000	-0.0001	0.3009	0.0870	0.0037	0.0092												
816	8	7	0.600	-0.01	-0.00	-0.02	-0.07	0.06	-0.03	0.6249	0.2911	0.0029	-0.0003	0.0001	-0.2086	0.5738	0.0015	0.0001	0.0000	0.6249	0.2911
		9	-0.0018	0.0000	-0.0001	0.3389	0.0012	-0.0000	0.0001	-0.1614	0.7603									-0.1614	0.7603
817	1	7	0.601	3.08	-0.00	-0.02	0.97	0.06	0.92	0.2672	0.6312	0.0006	0.0000	0.0002	0.0904	2.1127	0.0171	0.0009	0.0021	0.3423	0.6018
		9	-0.0006	-0.0000	-0.0003	0.1289	-0.0158	-0.0010	-0.0019												
817	2	7	0.599	-0.02	-0.00	-0.02	1.02	0.06	0.96	0.4061	0.6407	0.0022	-0.0000	0.0003	-0.0104	0.8861	0.0061	0.0005	0.0007	0.4061	0.6407
		9	-0.0018	0.0000	-0.0001	0.4613	0.0055	0.0001	0.0008	0.1681	0.7280									0.1681	0.7280
817	3	7	0.599	1.00	-0.00	-0.02	1.04	0.06	0.98	0.3284	0.6182	0.0019	0.0001	0.0004	0.2791	1.1945	0.0148	0.0009	0.0018	0.3284	0.6182
		9	-0.0017	0.0000	-0.0001	0.4225	0.0134	0.0006	0.0017	0.2291	0.6541									0.2291	0.6541
817	4	7	0.600	3.10	-0.00	-0.02	1.07	0.06	1.01	0.3103	0.6226	0.0018	0.0000	0.0003	0.1295	1.0397	0.0325	0.0020	0.0040	0.3103	0.6226
		9	-0.0023	0.0001	-0.0000	0.0767	0.0310	0.0016	0.0039	0.2720	0.6398									0.2720	0.6398
817	5	7	0.600	6.22	-0.00	-0.02	1.12	0.07	1.06	0.3032	0.5817	0.0014	0.0000	0.0002	0.2907	0.7681	0.0557	0.0033	0.0064	0.3032	0.5817
		9	-0.0003	0.0001	-0.0000	1.0654	0.0568	0.0031	0.0066	0.2779	0.5840									0.2779	0.5840
817	6	7	0.599	9.28	-0.00	-0.02	1.16	0.06	1.09	0.2721	0.5454	0.0001	0.0000	0.0002	-1.1155	-7.6385	0.0756	0.0041	0.0082	0.2721	0.5454
		9	-0.0010	-0.0000	-0.0003	-1.4887	0.0762	0.0038	0.0083	0.2541	0.5450									0.2541	0.5450
817	7	7	0.599	12.45	-0.00	-0.02	1.16	0.06	1.09	0.1991	0.5378	0.0006	0.0001	0.0001	-0.9482	1.5885	0.0878	0.0034	0.0094	0.1991	0.5378
		9	-0.0019	0.0000	-0.0002	0.5333	0.0890	0.0036	0.0095	0.2034	0.5340									0.2034	0.5340
817	8	7	0.600	-0.04	-0.00	-0.02	1.01	0.06	0.97	0.3793	0.6151	0.0018	0.0000	0.0004	0.1428	1.1755	0.0063	0.0004	0.0007	0.3793	0.6151
		9	-0.0016	0.0000	-0.0001	0.4626	0.0060	0.0002	0.0008	0.1783	0.7047									0.1783	0.7047

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key														
A18	1	7	0.599	-3.06	-0.001	-0.000	-0.001	-0.5377	-0.03	2.95	0.067	-0.0004	0.06	3.04	0.2122	0.6751
		8	0.0025	-0.0002	0.0001	0.0000	0.0001	-0.2459	0.1999	-0.0067	-0.0048	-0.0004	-0.0002	-0.0009	0.4595	0.5629
		9	-0.0028	0.0001	-0.0000	-0.0000	-0.0000	-0.0000	0.0666	-0.0048	-0.0048	-0.0004	-0.0004	-0.0005		
A18	2	7	0.600	-0.000	-0.000	-0.000	-0.0142	-0.02	3.00	0.0168	0.0161	0.0007	0.06	3.07	0.3225	0.6306
		8	0.0020	-0.0000	0.0003	0.0000	-0.2453	0.9666	0.2160	0.0168	0.0161	0.0007	0.0010	0.0021	0.2432	0.6497
		9	-0.0022	0.0001	-0.0000	-0.0000	-0.0000	0.2160	0.9666	0.0161	0.0161	0.0007	0.0010	0.0021		
A18	3	7	0.600	1.04	-0.000	-0.000	-0.0960	-0.02	3.01	0.255	0.0243	0.06	0.06	3.09	0.3095	0.6300
		8	0.0030	-0.0000	0.0003	0.0000	-0.2724	0.6195	0.1441	0.255	0.0243	0.06	0.0015	0.0032	0.2631	0.6420
		9	-0.0024	0.0001	-0.0000	-0.0000	-0.0000	0.1441	0.6195	0.0243	0.0243	0.06	0.0012	0.0031		
A18	4	7	0.600	3.13	-0.000	-0.000	0.6010	-0.02	3.05	0.438	0.0429	0.06	0.06	3.12	0.3006	0.6091
		8	0.0012	0.0001	0.0004	0.0000	0.3453	1.8794	0.2636	0.438	0.0429	0.06	0.0026	0.0053	0.2700	0.6225
		9	-0.0017	0.0001	-0.0000	-0.0000	-0.0000	0.2636	0.2636	0.0429	0.0429	0.06	0.0023	0.0053		
A18	5	7	0.600	6.21	-0.000	-0.000	-0.2638	-0.02	3.09	0.652	0.0662	0.07	0.07	3.16	0.2822	0.5685
		8	0.0009	-0.0000	0.0002	0.0000	-2.0619	1.2705	1.5836	0.652	0.0662	0.07	0.0036	0.0074	0.2695	0.5704
		9	-0.0002	0.0000	-0.0000	-0.0000	-0.0000	1.5836	1.5836	0.0662	0.0662	0.07	0.0035	0.0075		
A18	6	7	0.600	9.28	-0.000	-0.000	-0.4069	-0.02	3.12	0.813	0.0832	0.07	0.07	3.19	0.2458	0.5377
		8	0.0006	-0.0000	0.0002	0.0000	-0.4936	2.0470	2.1370	0.813	0.0832	0.07	0.0040	0.0087	0.2355	0.5377
		9	-0.0011	0.0001	-0.0000	-0.0000	-0.0000	2.1370	2.1370	0.0832	0.0832	0.07	0.0039	0.0089		
A18	7	7	0.599	12.46	-0.000	-0.000	-0.8459	-0.02	3.10	0.866	0.0893	0.06	0.06	3.18	0.1801	0.5358
		8	0.0016	-0.0002	0.0001	0.0001	-0.2647	0.3253	0.1257	0.866	0.0893	0.06	0.0031	0.0092	0.1846	0.5358
		9	-0.0042	0.0002	-0.0001	-0.0001	-0.0001	0.1257	0.1257	0.0893	0.0893	0.06	0.0032	0.0094		
A18	8	7	0.600	-0.02	-0.000	-0.000	-0.2035	-0.02	2.99	0.170	0.0157	0.06	0.06	3.07	0.3171	0.6181
		8	0.0027	-0.0001	0.0003	0.0003	-0.2272	0.6224	0.2616	0.170	0.0157	0.06	0.0010	0.0021	0.2551	0.6181
		9	-0.0022	0.0001	-0.0001	-0.0001	-0.0001	0.2616	0.2616	0.0157	0.0157	0.06	0.0008	0.0021		
A19	1	7	0.600	-3.05	-0.000	-0.000	-0.3456	-0.02	3.00	0.064	0.0072	0.06	0.06	6.01	0.4010	0.7663
		8	0.0018	-0.0001	0.0002	0.0002	-0.2435	0.6378	0.3208	0.064	0.0072	0.06	0.0005	0.0008	0.2129	0.7663
		9	-0.0019	0.0000	-0.0001	-0.0001	-0.0001	0.3208	0.3208	0.0072	0.0072	0.06	0.0003	0.0011		
A19	2	7	0.600	0.00	-0.000	-0.000	-0.0892	-0.02	3.05	0.330	0.0327	0.06	0.06	6.06	0.3105	0.6462
		8	0.0023	-0.0000	0.0003	0.0003	-0.1051	0.8315	1.3688	0.330	0.0327	0.06	0.0020	0.0042	0.2617	0.6462
		9	-0.0009	0.0000	-0.0002	-0.0002	-0.0002	1.3688	1.3688	0.0327	0.0327	0.06	0.0017	0.0042		
A19	3	7	0.600	1.03	-0.000	-0.000	-0.0390	-0.02	6.07	0.424	0.0418	0.06	0.06	6.07	0.3044	0.6260
		8	0.0025	-0.0000	0.0003	0.0003	-0.1023	0.7575	0.8067	0.424	0.0418	0.06	0.0025	0.0053	0.2704	0.6260
		9	-0.0015	0.0000	-0.0002	-0.0002	-0.0002	0.8067	0.8067	0.0418	0.0418	0.06	0.0022	0.0053		

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key →														
A20	7	0.601 0.0000 0.0002	12.44 -0.0000 -0.0001	-0.000 0.0002 -0.0004	-2.3170 -7.8487	-0.02	9.18 0.0867 0.0893	0.06 0.0024 0.0024	9.09 0.0095 0.0097	0.1425 0.1354	0.5507 0.5439						
A20	8	0.600 0.0024 -0.0017	0.01 -0.0000 0.0000	-0.000 0.0003 -0.0001	-0.1703 0.7463 0.4836	-0.02 0.0489 0.0487	9.15 0.0489 0.0487	0.06 0.0029 0.0026	9.07 0.0059 0.0060	0.3032 0.2714	0.6059 0.6192						
A21	1	0.599 0.0002 -0.0007	-3.00 0.0001 0.0000	-0.000 0.0003 -0.0002	1.9188 -0.1735	-0.02 7.3886 1.6916	15.09 0.0530 0.0546	0.06 0.0031 0.0029	15.09 0.0064 0.0067	0.2926 0.2690	0.6040 0.6156						
A21	2	0.600 0.0021 -0.0005	0.01 -0.0000 -0.0000	-0.000 0.0004 -0.0002	-0.0403 0.0200	-0.02 0.9397 2.3159	15.14 0.0748 0.0748	0.06 0.0040 0.0038	15.12 0.0085 0.0085	0.2679 0.2567	0.5685 0.5688						
A21	3	0.599 0.0022 -0.0010	1.06 0.0000 -0.0000	-0.000 0.0004 -0.0002	0.0612 0.0274	-0.02 0.9557 1.3159	15.14 0.0793 0.0796	0.06 0.0040 0.0039	15.13 0.0087 0.0088	0.2572 0.2449	0.5539 0.5538						
A21	4	0.599 0.0021 -0.0011	3.18 0.0000 0.0000	-0.000 0.0004 -0.0001	0.0156 -0.3233	-0.02 0.9348 0.5533	15.13 0.0850 0.0864	0.06 0.0035 0.0037	15.13 0.0092 0.0093	0.2109 0.2166	0.5452 0.5430						
A21	5	0.599 0.0022 -0.0030	6.23 -0.0002 0.0003	-0.000 0.0001 0.0002	-0.5578 -0.5093	-0.03 0.3275 -0.4341	15.12 0.0875 0.0897	0.06 0.0029 0.0031	15.12 0.0095 0.0096	0.1666 0.1743	0.5462 0.5366						
A21	6	0.600 0.0007 -0.0019	9.31 -0.0001 0.0001	-0.000 0.0002 0.0000	-0.7087 -0.3368	-0.02 1.5634 -0.0318	15.12 0.0889 0.0911	0.07 0.0025 0.0024	15.11 0.0097 0.0099	0.1418 0.1356	0.5502 0.5460						
A21	7	0.600 0.0007 -0.0016	12.48 -0.0001 0.0000	-0.000 0.0002 -0.0002	-0.9375 -0.0409	-0.02 1.6211 0.6158	15.12 0.0939 0.0957	0.06 0.0024 0.0023	15.11 0.0103 0.0105	0.1306 0.1241	0.5522 0.5503						
A21	8	0.601 0.0021 -0.0013	0.02 -0.0000 0.0000	-0.000 0.0004 -0.0001	-0.0133 -0.1718	-0.02 1.0101 0.6926	15.13 0.0752 0.0754	0.06 0.0040 0.0038	15.13 0.0085 0.0085	0.2671 0.2548	0.5668 0.5684						

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																					
A23	4	7	0.902	3.13	-0.0004	-0.0912	-0.02	-0.01	0.05	0.00	0.0041	0.2156	0.6490	0.0032	-0.0000	0.0004	-0.0912	0.7621	0.0322	0.0013	0.0041	0.1857	0.6583	
		8	-0.0001	0.0001	-0.0003	-4.6845	11.0416	0.0294	0.0010	0.0038	0.0038													
A23	5	7	0.899	6.19	-0.0000	-0.2026	-0.02	0.02	0.05	0.05	0.0071	0.2122	0.6112	0.0006	-0.0000	0.0000	-0.2026	0.2137	0.0582	0.0024	0.0071	0.1939	0.6153	
		8	0.0001	0.0002	-0.0001	9.6944	-5.4465	0.0595	0.0023	0.0073	0.0073													
A23	6	7	0.899	9.31	-0.0000	-0.0773	-0.02	0.06	0.04	0.08	0.0093	0.1897	0.5814	0.0505	-0.0000	0.0001	-0.0773	1.4519	0.0805	0.0030	0.0093	0.1854	0.5756	
		8	0.0011	0.0001	-0.0002	0.5008	-1.2491	0.0817	0.0030	0.0094	0.0094													
A23	7	7	0.900	9.31	-0.0000	-0.3032	-0.02	0.05	0.04	0.07	0.0093	0.1906	0.5827	0.0009	-0.0000	0.0001	-0.3032	0.7280	0.0803	0.0030	0.0093	0.1865	0.5751	
		8	0.0004	0.0002	-0.0001	-2.5635	1.5105	0.0815	0.0030	0.0093	0.0093													
A23	8	7	0.901	12.47	-0.0000	-0.5965	-0.02	0.06	0.05	0.09	0.0110	0.1544	0.5677	0.0020	-0.0000	0.0002	-0.5965	0.4925	0.0975	0.0030	0.0110	0.1522	0.5627	
		8	0.0014	0.0002	-0.0001	-0.7963	0.5423	0.0985	0.0030	0.0030	0.0110													
A23	9	7	0.901	-0.01	-0.0000	-0.1408	-0.02	-0.06	0.04	-0.02	-0.0001	-2.2673	-0.3120	0.0040	-0.0000	0.0005	-0.1408	0.6814	0.0021	0.0001	-0.0001	-2.2673	-0.3120	
		8	0.0009	0.0002	-0.0003	-1.1604	1.7349	0.0001	0.0001	0.0001	0.0001													
A24	1	7	0.898	-0.02	-0.0000	-0.0845	-0.01	1.01	0.04	0.97	0.0008	0.3436	0.6190	0.0036	-0.0000	0.0005	-0.0845	0.8123	0.0068	0.0004	0.0008	0.1177	0.6723	
		8	0.0007	0.0001	-0.0003	-1.3297	2.4214	0.0060	0.0001	0.0008	0.0008													
A24	2	7	0.897	-3.05	-0.0000	-0.2597	-0.02	0.97	0.04	0.93	0.0028	0.2125	0.6841	0.0030	-0.0000	0.0003	-0.2597	0.5596	-0.0208	0.0008	0.0028	0.2817	0.6574	
		8	0.0026	0.0003	-0.0003	-0.5832	0.5768	-0.0196	-0.0011	-0.0025	-0.0025													
A24	3	7	0.897	-0.02	-0.0000	-0.1172	-0.02	1.01	0.04	0.98	0.0008	0.3506	0.6448	0.0036	-0.0000	0.0003	-0.1172	0.7627	0.0067	0.0004	0.0008	0.1131	0.6737	
		8	0.0006	0.0001	-0.0003	-1.5031	2.5442	0.0062	0.0001	0.0008	0.0008													
A24	4	7	0.900	1.00	-0.0000	-0.1406	-0.02	1.03	0.04	0.99	0.0023	0.2184	0.6563	0.0044	-0.0000	0.0006	-0.1406	0.6828	0.0177	0.0007	0.0023	0.1695	0.6710	
		8	0.0007	0.0001	-0.0003	-1.1496	2.0491	0.0150	0.0005	0.0020	0.0020													
A24	5	7	0.899	3.13	-0.0000	-0.0618	-0.02	1.06	0.04	1.02	0.0050	0.2118	0.6451	0.0029	-0.0000	0.0005	-0.0618	0.8346	0.0391	0.0016	0.0050	0.1909	0.6474	
		8	0.0010	0.0002	-0.0002	-1.1790	1.0720	0.0360	0.0013	0.0046	0.0046													

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																		
824	6	0.097	6.21	-0.000	-0.000	-0.02	1.10	0.04	1.06	0.2103	0.6016	0.0010	-0.0003	0.0000	-0.1515	0.0634	0.0025	0.0076	0.1993	0.6029
824	7	0.098	9.30	-0.001	-0.001	-0.02	1.13	0.04	1.08	0.1792	0.5754	0.0011	-0.0002	0.0001	-0.3344	0.0851	0.0031	0.0097	0.1832	0.5759
824	8	0.097	12.46	-0.001	-0.001	-0.02	1.14	0.04	1.09	0.1413	0.5680	0.0017	-0.0002	0.0003	-0.4059	0.1013	0.0029	0.0113	0.1449	0.5642
824	9	0.090	-0.01	-0.005	-0.002	-0.02	1.01	0.04	0.98	0.3117	0.5921	0.0045	-0.0002	0.0005	-0.5564	0.0062	0.0001	0.0008	0.1153	0.6658
825	3	0.026	3.05	-0.003	-0.002	-0.02	2.95	0.05	3.02	0.1286	0.6458	0.0026	-0.0001	0.0003	-0.7220	0.0084	0.0002	0.0010	0.3039	0.6789
825	4	0.044	-0.001	-0.001	-0.001	-0.02	3.01	0.05	3.06	0.2674	0.6859	0.0044	-0.0001	0.0005	-0.6127	0.0193	0.0010	0.0025	0.1796	0.6840
825	5	0.044	1.05	-0.006	-0.002	-0.02	3.02	0.05	3.08	0.2413	0.6678	0.0044	-0.0001	0.0006	-0.7337	0.0288	0.0011	0.0038	0.1926	0.6609
825	6	0.029	3.12	-0.004	-0.002	-0.02	3.05	0.05	3.11	0.2269	0.6311	0.0029	-0.0001	0.0003	-0.6845	0.0492	0.0019	0.0063	0.1954	0.6421
825	7	0.007	6.21	-0.000	-0.000	-0.02	3.09	0.05	3.14	0.2111	0.5903	0.0007	-0.0002	0.0000	-0.6731	0.0728	0.0030	0.0084	0.1949	0.5896
825	8	0.019	9.30	-0.001	-0.002	-0.02	3.11	0.05	3.16	0.1806	0.5694	0.0019	-0.0002	0.0001	-0.5539	0.0922	0.0031	0.0101	0.1691	0.5603
825	9	0.009	12.47	-0.001	-0.001	-0.02	3.11	0.05	3.16	0.1304	0.5738	0.0009	-0.0001	0.0002	-0.958	0.1070	0.0026	0.0118	0.1273	0.5595

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																				
825	10	7	0.899	0.0001	-0.0005	-0.1593	-0.02	3.00	0.05	3.06	0.2527	0.6663	0.0042	-0.0002	0.0005	-0.1593	-0.02	3.00	0.05	3.06	0.2527	0.6663	
		6	0.0019	0.0002	0.0001	0.0001	0.3438	0.0199	0.0010	0.0026	0.1695	0.6663	0.0019	0.0002	0.0001	0.0001	0.3438	0.0199	0.0010	0.0026	0.1695	0.6663	0.6663
		9	-0.0014	0.0001	-0.0003	-0.02178	1.4141	0.0080	0.0006	0.0011	0.3939	0.7474	-0.0014	0.0001	-0.0003	-0.02178	1.4141	0.0080	0.0006	0.0011	0.3939	0.7474	0.7474
826	1	7	0.899	0.0001	-0.0005	-0.1593	-0.02	3.00	0.05	3.06	0.2527	0.6663	0.0042	-0.0002	0.0005	-0.1593	-0.02	3.00	0.05	3.06	0.2527	0.6663	
		6	0.0013	0.0001	0.0002	0.2390	0.8839	0.0382	0.0018	0.0052	0.2335	0.6628	0.0013	0.0001	0.0002	0.2390	0.8839	0.0382	0.0018	0.0052	0.2335	0.6628	0.6628
		9	-0.0013	0.0001	-0.0002	-0.7039	0.8839	0.0382	0.0015	0.0050	0.2007	0.6599	-0.0013	0.0001	-0.0002	-0.7039	0.8839	0.0382	0.0015	0.0050	0.2007	0.6599	0.6599
826	3	7	0.898	1.04	-0.0006	-0.0678	-0.01	6.08	0.05	6.06	0.2312	0.6364	0.0037	-0.0001	0.0006	-0.0678	-0.01	6.08	0.05	6.06	0.2312	0.6364	
		6	0.0009	0.0001	-0.0002	-0.8737	1.1840	0.0477	0.0022	0.0061	0.1975	0.6406	0.0009	0.0001	-0.0002	-0.8737	1.1840	0.0477	0.0022	0.0061	0.1975	0.6406	0.6406
		9	-0.0009	0.0001	-0.0002	0.2291	1.1067	0.0640	0.0028	0.0077	0.2205	0.6031	-0.0009	0.0001	-0.0002	0.2291	1.1067	0.0640	0.0028	0.0077	0.2205	0.6031	0.6031
826	4	7	0.900	3.12	-0.0003	-1.0104	0.02	6.10	0.05	6.09	0.2205	0.5989	0.0016	0.0002	0.0001	-1.0104	0.02	6.10	0.05	6.09	0.2205	0.5989	
		6	0.0012	0.0002	-0.0001	0.2291	1.1067	0.0640	0.0028	0.0077	0.2205	0.6031	0.0012	0.0002	-0.0001	0.2291	1.1067	0.0640	0.0028	0.0077	0.2205	0.6031	0.6031
		9	-0.0012	0.0002	-0.0001	-1.0104	0.7127	0.0636	0.0025	0.0076	0.2012	0.5989	-0.0012	0.0002	-0.0001	-1.0104	0.7127	0.0636	0.0025	0.0076	0.2012	0.5989	0.5989
826	5	7	0.902	6.22	-0.0002	-0.1632	-0.02	6.13	0.05	6.11	0.1955	0.5766	0.0009	-0.0003	0.0000	-0.1632	-0.02	6.13	0.05	6.11	0.1955	0.5766	
		6	0.0004	0.0003	-0.0000	-3.1712	0.1438	0.0823	0.0031	0.0092	0.1888	0.5645	0.0004	0.0003	-0.0000	-3.1712	0.1438	0.0823	0.0031	0.0092	0.1888	0.5645	0.5645
		9	-0.0004	0.0003	-0.0000	1.7328	-2.7125	0.0997	0.0029	0.0111	0.1494	0.5712	-0.0004	0.0003	-0.0000	1.7328	-2.7125	0.0997	0.0029	0.0111	0.1494	0.5712	0.5712
826	6	7	0.699	12.46	-0.0001	-0.9580	-0.02	6.13	0.05	6.12	0.1161	0.5730	0.0009	-0.0001	0.0000	-0.9580	-0.02	6.13	0.05	6.12	0.1161	0.5730	
		6	0.0014	0.0002	-0.0001	-0.7195	0.6033	0.1113	0.0024	0.0122	0.1166	0.5730	0.0014	0.0002	-0.0001	-0.7195	0.6033	0.1113	0.0024	0.0122	0.1166	0.5730	0.5730
		9	-0.0014	0.0002	-0.0001	0.8661	0.9148	0.0379	0.0015	0.0051	0.2302	0.6615	-0.0014	0.0002	-0.0001	0.8661	0.9148	0.0379	0.0015	0.0051	0.2302	0.6615	0.6615
826	7	7	0.898	-3.05	-0.0004	-0.2838	-0.02	9.07	0.04	9.01	0.2605	0.6834	0.0040	-0.0001	0.0000	-0.2838	-0.02	9.07	0.04	9.01	0.2605	0.6834	
		6	0.0025	0.0002	-0.0003	-0.5419	0.7408	0.0271	0.0011	0.0036	0.2037	0.6781	0.0025	0.0002	-0.0003	-0.5419	0.7408	0.0271	0.0011	0.0036	0.2037	0.6781	0.6781
		9	-0.0025	0.0002	-0.0003	0.2155	1.0630	0.0549	0.0021	0.0067	0.2310	0.6207	-0.0025	0.0002	-0.0003	0.2155	1.0630	0.0549	0.0021	0.0067	0.2310	0.6207	0.6207
827	1	7	0.898	0.04	-0.0004	-0.2155	-0.02	9.12	0.04	9.05	0.2310	0.6207	0.0040	-0.0001	0.0000	-0.2155	-0.02	9.12	0.04	9.05	0.2310	0.6207	
		6	0.0040	-0.0001	0.0004	-0.8220	1.0630	0.0549	0.0021	0.0067	0.2310	0.6207	0.0040	-0.0001	0.0004	-0.8220	1.0630	0.0549	0.0021	0.0067	0.2310	0.6207	0.6207
		9	-0.0040	-0.0001	-0.0002	0.8220	1.0630	0.0549	0.0021	0.0067	0.2310	0.6207	-0.0040	-0.0001	-0.0002	0.8220	1.0630	0.0549	0.0021	0.0067	0.2310	0.6207	0.6207

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																			
A2A	6	7	0.898	9.32	-0.0002	-0.6017	1.4148	15.13	0.05	15.12	0.1135	0.5657	0.0008	-0.0001	-0.0002	-2.0186	1.4148	15.13	0.05	15.12	0.1135	0.5657
A2A	7	6	0.899	12.49	-0.0001	-0.5291	0.4505	15.14	0.05	15.11	0.0963	0.5706	0.0021	-0.0001	-0.0001	-0.7155	0.4505	15.14	0.05	15.11	0.0963	0.5706
A2A	8	9	0.899	0.05	-0.0003	-0.1696	0.9536	15.12	0.05	15.10	0.1994	0.5723	0.0021	-0.0002	-0.0002	-0.2622	0.9536	15.12	0.05	15.10	0.1994	0.5723
A29	3	7	0.898	3.14	44.99	0.2907	0.6197	-3.00	-3.04	-3.02	0.2409	0.6510	0.358	-0.0044	0.2907	0.6197	-3.00	-3.04	-3.02	0.2409	0.6510	
A29	4	6	0.899	0.03	44.99	0.2754	0.5450	-2.97	-3.01	-2.99	0.1941	0.6837	0.156	-0.0017	0.2754	0.5450	-2.97	-3.01	-2.99	0.1941	0.6837	
A29	5	9	0.898	1.00	44.99	0.3935	0.6646	-2.99	-2.99	-2.98	0.1775	0.7434	0.077	-0.0008	0.3935	0.6646	-2.99	-2.99	-2.98	0.1775	0.7434	
A29	6	7	0.896	3.06	44.99	-0.0730	0.7420	-2.93	-2.97	-2.97	1.0100	0.4193	0.052	-0.0007	-0.0730	0.7420	-2.93	-2.97	-2.97	1.0100	0.4193	
A29	7	6	0.896	6.16	44.99	0.1413	0.6295	-2.91	-2.92	-2.93	0.1587	0.6505	0.262	-0.0033	0.1413	0.6295	-2.91	-2.92	-2.93	0.1587	0.6505	
A29	8	9	0.897	9.31	44.99	0.1590	0.6491	-2.87	-2.89	-2.90	0.1614	0.6298	0.479	-0.0061	0.1590	0.6491	-2.87	-2.89	-2.90	0.1614	0.6298	
A29	9	7	0.897	12.42	45.00	0.1337	0.6360	-2.84	-2.86	-2.87	0.1467	0.6236	0.710	-0.0090	0.1337	0.6360	-2.84	-2.86	-2.87	0.1467	0.6236	
A29	10	6	0.895	-0.03	44.99	0.3389	0.6122	-3.01	-3.01	-2.99	0.2118	0.7091	0.574	-0.0034	0.3389	0.6122	-3.01	-3.01	-2.99	0.2118	0.7091	
		9	0.895	-0.0007	-0.0018	0.1956	0.6695	-0.0153	-0.0009	-0.0021	0.3192	0.7086	0.148	-0.0024	0.1956	0.6695	-0.0153	-0.0009	-0.0021	0.3192	0.7086	

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																		
A30	1	7	0.897	-3.09	44.99	0.2742	-0.598	-0.09	0.02	-0.06	0.2300	0.6793	0.0184	-0.0010	-0.0020	0.0020	0.0008	-0.0025	0.2548	0.6751	
		8	-0.0209	-0.0007	-0.0027	0.1712	0.6662	-0.0180	-0.0009	-0.0008	-0.0024										
		9																			
A30	2	7	0.897	-0.0002	44.99	-0.4943	-0.909	-0.002	0.05	-0.001	-0.8689	2.0206	0.0020	-0.0002	0.0003	-0.0000	0.0000	-0.0001	-1.5213	0.9831	
		8	-0.0012	-0.0002	-0.0000	-1.0998	0.2197	-0.0002	-0.0000	-0.0000	-0.0000										
		9																			
A30	3	7	0.897	-0.0001	44.99	-0.4055	-0.2195	-0.004	0.05	-0.001	-0.5082	1.6699	0.0017	-0.0001	0.0004	-0.0000	0.0000	-0.0001	-0.9777	0.7948	
		8	-0.0007	-0.0002	-0.0001	-1.6484	0.9331	-0.0004	-0.0000	-0.0000	-0.0000										
		9																			
A30	4	7	0.898	1.03	44.99	0.0777	-0.001	-0.05	0.07	-0.003	0.4275	0.6085	0.0088	0.0001	0.0003	0.0001	0.0008	0.1222	0.6924		
		8	0.0053	0.0005	0.0012	0.5489	0.5672	0.0061	0.0001	0.0001	0.0001										
		9																			
A30	5	7	0.895	3.09	44.99	0.1537	0.429	-0.03	0.10	-0.03	0.1908	0.6811	0.0249	0.0007	0.0007	0.0006	0.0024	0.1879	0.6626		
		8	0.0185	0.0014	0.0032	0.3919	0.6958	0.0186	0.0006	0.0006	0.0024										
		9																			
A30	6	7	0.896	6.21	44.99	0.1662	0.641	-0.00	0.14	-0.00	0.1507	0.6566	0.0470	0.0015	0.0013	0.0018	0.0049	0.2341	0.6355		
		8	0.0416	0.0024	0.0062	0.2955	0.6178	0.0390	0.0018	0.0018	0.0049										
		9																			
A30	7	7	0.896	9.33	44.99	0.1410	0.607	0.02	0.16	0.02	0.4926	0.6226	0.0687	0.0019	0.0019	0.0027	0.0079	0.4370	0.5881		
		8	0.0564	0.0033	0.0067	0.2962	0.5999	0.0576	0.0027	0.0027	0.0067										
		9																			
A30	8	7	0.899	12.44	45.00	0.1323	0.6145	0.05	0.18	0.05	0.1337	0.6206	0.0868	0.0022	0.00106	0.0022	0.0080	0.2286	0.5670		
		8	0.0709	0.0037	0.0080	0.2625	0.5657	0.0713	0.0032	0.0032	0.0080										
		9																			
A30	9	7	0.898	-0.001	44.99	-0.3545	-0.002	-0.06	0.04	-0.06	-1.4961	3.5357	0.0118	-0.0003	-0.0000	-0.0000	-0.0001	-1.5586	2.2801		
		8	-0.0013	-0.0003	-0.0000	-1.1666	0.1752	-0.0002	-0.0000	-0.0000	-0.0001										
		9																			
A31	1	7	0.897	-3.08	44.99	0.3292	0.5872	-0.0130	0.99	-0.005	0.1977	0.6975	-0.0115	-0.0007	-0.0013	-0.0017	-0.0017	0.2626	0.6931		
		8	-0.0160	-0.0003	-0.0018	0.1119	0.5911	-0.0123	-0.0006	-0.0006	-0.0017										
		9																			
A31	2	7	0.899	-0.00	44.99	0.1364	0.7990	1.01	1.02	0.003	0.4139	0.5964	0.0070	0.0001	0.0011	0.0005	0.0007	0.1263	0.6789		
		8	0.0027	0.0006	0.0005	1.0264	0.9401	0.0052	0.0001	0.0001	0.0005										
		9																			

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key													
A31	3	0.898 0.0162 0.0102	1.04 0.0004 0.0009	44.99 0.0021 0.0012	0.1300 0.4549	1.04 0.6481 0.6361	1.02 0.0111 0.0112	1.04 0.0005 0.0004	0.97 0.0014 0.0015	0.2614 0.1911	0.6670 0.6716				
A31	4	0.896 0.0321 0.0257	3.10 0.0011 0.0016	44.99 0.0042 0.0033	0.1734 0.3282	1.06 0.6575 0.6504	1.04 0.0262 0.0244	1.07 0.0009 0.0010	1.00 0.0035 0.0031	0.1755 0.2064	0.6687 0.6457				
A31	5	0.898 0.0537 0.0464	6.22 0.0017 0.0026	44.99 0.0070 0.0055	0.1593 0.2900	1.09 0.6581 0.6014	1.07 0.0504 0.0450	1.10 0.0015 0.0021	1.03 0.0065 0.0056	0.1522 0.2340	0.6525 0.6503				
831	6	0.899 0.0745 0.0608	9.35 0.0020 0.0035	44.99 0.0092 0.0071	0.1357 0.2881	1.12 0.6226 0.5835	1.10 0.0700 0.0618	1.13 0.0020 0.0029	1.05 0.0086 0.0072	0.1449 0.2391	0.6213 0.5869				
831	7	0.900 0.0924 0.0733	12.45 0.0024 0.0037	45.00 0.0114 0.0081	0.1298 0.2558	1.14 0.6167 0.5519	1.12 0.0900 0.0753	1.15 0.0023 0.0033	1.07 0.0112 0.0084	0.1287 0.2191	0.6210 0.5614				
A31	8	0.897 0.0075 0.0028	-0.00 0.0001 0.0006	44.99 0.0010 0.0005	0.1036 1.0823	1.02 0.7177 0.9095	1.01 0.0047 0.0054	1.02 0.0003 0.0001	0.96 0.0005 0.0007	0.3975 0.1336	0.5998 0.6514				
832	1	0.897 -0.0002 -0.0039	-3.06 -0.0001 0.0001	44.99 0.0002 -0.0004	4.0713 -0.2359	3.01 -5.1555 0.5136	2.97 -0.0012 -0.0017	2.90 0.0000 -0.0000	3.05 -0.0002 -0.0002	-0.0657 -0.231A	0.6816 0.6820				
832	2	0.897 0.0206 0.0148	0.03 0.000A 0.0011	44.99 0.0028 0.0019	0.1993 0.4038	3.04 0.6811 0.6548	3.00 0.0175 0.0185	2.93 0.0008 0.0006	3.08 0.0024 0.0025	0.2424 0.1874	0.6840 0.6796				
832	3	0.897 0.0290 0.0231	1.06 0.0011 0.0015	44.99 0.0039 0.0029	0.2025 0.3350	3.05 0.6727 0.6484	3.01 0.0249 0.0248	2.94 0.0010 0.0010	3.09 0.0033 0.0033	0.2122 0.2084	0.6670 0.6627				
832	4	0.896 0.0457 0.0369	3.11 0.0015 0.0023	44.99 0.0060 0.0048	0.1725 0.3128	3.08 0.6658 0.6496	3.03 0.0406 0.0378	2.98 0.0013 0.0016	3.12 0.0054 0.0048	0.1650 0.2192	0.6693 0.6469				
A32	5	0.897 0.0661 0.0552	6.24 0.0020 0.0032	44.99 0.0083 0.0066	0.1538 0.2943	3.11 0.6310 0.6012	3.06 0.0612 0.0550	3.01 0.0019 0.0025	3.14 0.0077 0.0066	0.1565 0.2354	0.6291 0.6087				

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key											
A32	6	7	0.896	9.37	44.99	0.1369	3.13	3.09	3.03	3.16	0.1425	0.6157		
			0.0844	0.0023	0.0102	0.2745	0.6077	0.0810	0.0023	0.0099	0.2305	0.5698		
			0.0692	0.0038	0.0080	0.0000	0.5806	0.0705	0.0032	0.0080				
A32	7	8	0.898	12.47	45.00	0.1260	3.15	3.11	3.04	3.16	0.1281	0.6126		
			0.1020	0.0025	0.0124	0.2314	0.6084	0.0994	0.0025	0.0121	0.2013	0.5510		
			0.0781	0.0036	0.0086	0.0000	0.5554	0.0810	0.0032	0.0089				
A32	8	9	0.896	0.02	44.99	0.1963	3.04	3.00	2.93	3.08	0.2341	0.6695		
			0.0206	0.0008	0.0028	0.4746	0.6803	0.0177	0.0008	0.0023	0.1869	0.6775		
			0.0135	0.0012	0.0020	0.0000	0.7531	0.0185	0.0006	0.0025				
A33	1	7	0.896	-3.05	44.99	0.2279	6.03	6.02	6.07	6.01	0.2837	0.7163		
			0.0174	0.0007	0.0023	0.4284	0.6768	0.0156	0.0008	0.0022	0.1648	0.6995		
			0.0129	0.0011	0.0016	0.0000	0.6436	0.0167	0.0005	0.0023				
A33	2	7	0.899	0.06	44.99	0.1929	6.07	6.06	6.11	6.03	0.2063	0.6712		
			0.0415	0.0016	0.0055	0.3094	0.6647	0.0387	0.0015	0.0051	0.2119	0.6627		
			0.0349	0.0021	0.0047	0.0000	0.6754	0.0371	0.0015	0.0049				
A33	3	7	0.898	1.10	44.99	0.1971	6.08	6.06	6.12	6.04	0.1871	0.6599		
			0.0492	0.0019	0.0067	0.2875	0.6807	0.0459	0.0017	0.0060	0.2099	0.6396		
			0.0424	0.0024	0.0052	0.0000	0.6231	0.0439	0.0018	0.0056				
A33	4	7	0.899	3.16	44.99	0.1721	6.10	6.08	6.14	6.05	0.1737	0.6303		
			0.0622	0.0021	0.0077	0.2901	0.6243	0.0579	0.0020	0.0073	0.2207	0.6224		
			0.0538	0.0031	0.0065	0.0000	0.6107	0.0543	0.0023	0.0067				
A33	5	7	0.898	6.26	44.99	0.1496	6.12	6.11	6.17	6.08	0.1554	0.6130		
			0.0807	0.0024	0.0097	0.2749	0.6069	0.0768	0.0023	0.0094	0.2307	0.5769		
			0.0689	0.0037	0.0080	0.0000	0.5854	0.0678	0.0031	0.0078				
A33	6	7	0.897	9.40	44.99	0.1358	6.14	6.12	6.18	6.09	0.1397	0.6030		
			0.0977	0.0026	0.0117	0.2400	0.6009	0.0950	0.0026	0.0114	0.2057	0.5531		
			0.0788	0.0037	0.0088	0.0000	0.5612	0.0816	0.0033	0.0090				
A33	7	7	0.897	12.49	45.00	0.1245	6.16	6.14	6.17	6.09	0.1272	0.6008		
			0.1141	0.0028	0.0136	0.1827	0.5972	0.1110	0.0028	0.0133	0.1682	0.5505		
			0.0866	0.0031	0.0096	0.0000	0.5567	0.0903	0.0030	0.0099				
A33	8	7	0.899	0.05	44.99	0.2065	6.07	6.06	6.11	6.03	0.2031	0.6617		
			0.0404	0.0016	0.0055	0.3082	0.6822	0.0389	0.0015	0.0051	0.2122	0.6579		
			0.0349	0.0021	0.0046	0.0000	0.6678	0.0372	0.0015	0.0048				

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key															
836	11	1.249	12.60	-0.001	-0.3568	-0.092	-2.78	0.06	-2.83	0.1041	0.6336	0.0012	0.0000	0.0018	0.0112	0.0945	0.6307
		0.0013	-0.0003	0.0000	-1.2133	0.5433	0.0886	0.0016	0.0112								
		-0.0013	0.0003	0.0000	-1.2133	-0.0982	0.0890	0.0016	0.0112								
836	12	1.250	-0.001	0.0005	-0.2737	-0.3092	-2.97	0.06	-2.98	0.1323	0.7388	0.0020	0.0001	-0.0003	0.0021	0.2248	0.7042
		0.0033	-0.0001	0.0005	-0.2737	0.7639	-0.0141	-0.0006	-0.0020								
		-0.0018	0.0002	-0.0001	-0.7747	0.3092	-0.0155	-0.0006	-0.0021								
837	1	1.251	-3.18	-0.003	-0.3929	-0.9831	-0.11	0.05	-0.07	0.1125	0.6828	0.0038	0.0002	-0.0003	0.0036	0.1629	0.6765
		0.0017	-0.0001	0.0003	-0.3929	0.9831	-0.0279	-0.0006	-0.0038								
		-0.0014	0.0002	-0.0002	-0.7679	0.9732	-0.0268	-0.0006	-0.0036								
837	2	1.250	-0.001	0.0004	-0.2739	-0.7491	-0.06	0.04	-0.02	0.7648	0.6640	0.0001	0.0000	0.0001	0.0000	0.7648	0.6640
		0.0029	-0.0001	0.0004	-0.2739	0.7491	0.0007	0.0001	-0.0002								
		-0.0009	0.0002	-0.0002	-1.1197	1.4244	0.0001	-0.0002	0.0000	-6.3857	1.6430	0.0000	0.0000	0.0000	0.0000	-6.3857	1.6430
837	3	1.252	0.99	-0.005	-0.1667	-0.8186	-0.04	0.05	-0.01	0.1612	0.6576	0.0012	0.0000	0.0003	0.0010	0.0074	0.6783
		0.0030	-0.0001	0.0005	-0.1667	0.8186	0.0097	0.0003	-0.0012								
		-0.0008	0.0002	-0.0001	-1.4053	0.9890	0.0080	0.0000	0.0010								
837	4	1.247	3.14	-0.003	-0.2023	-0.6518	-0.00	0.05	0.00	0.1183	0.6581	0.0039	0.0000	0.0004	0.0036	0.0824	0.6535
		0.0030	-0.0001	0.0003	-0.2023	0.6518	0.0299	0.0007	0.0039								
		-0.0003	0.0002	-0.0000	-3.0913	0.7363	0.0276	0.0004	0.0036								
837	5	1.248	6.28	-0.002	-0.4792	-0.6437	0.03	0.05	0.05	0.1123	0.6614	0.0074	0.0000	0.0012	0.0074	0.0970	0.6545
		0.0016	-0.0001	0.0002	-0.4792	0.6437	0.0562	0.0012	0.0074								
		0.0012	0.0003	0.0001	1.1909	0.7708	0.0572	0.0011	0.0074								
837	6	1.249	9.41	-0.002	-0.3116	-0.7421	0.07	0.06	0.08	0.1047	0.6432	0.0104	0.0000	0.0016	0.0103	0.0949	0.6380
		0.0016	-0.0001	0.0002	-0.3116	0.7421	0.0809	0.0016	0.0104								
		-0.0007	0.0002	-0.0000	-1.9915	0.2376	0.0814	0.0015	0.0103								
837	7	1.251	12.66	-0.001	-0.3838	-0.4331	0.10	0.05	0.10	0.0950	0.6247	0.0127	0.0000	0.0019	0.0126	0.0876	0.6196
		0.0013	-0.0001	0.0001	-0.3838	0.4331	0.1017	0.0019	0.0127								
		-0.0011	0.0003	0.0000	-1.4958	-0.0510	0.1022	0.0017	0.0126								
837	8	1.249	-0.001	0.0004	-0.3062	-0.7133	-0.06	0.05	-0.02	0.4762	0.3837	0.0000	0.0000	0.0000	0.0000	0.4762	0.3837
		0.0031	-0.0001	0.0004	-0.3062	0.7133	0.0008	0.0000	-0.0002								
		-0.0013	0.0002	-0.0001	-1.0024	0.7149	0.0000	-0.0002	-0.0002								
83A	1	1.250	-3.13	-0.003	-0.4191	-0.8746	0.96	0.04	0.92	0.1186	0.6795	0.0028	0.0000	0.0004	0.0027	0.1186	0.6785
		0.0019	-0.0001	0.0003	-0.4191	0.8746	-0.0204	-0.0004	-0.0028								
		-0.0018	0.0002	-0.0002	-0.6813	0.6850	-0.0204	-0.0007	-0.0027								

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key												
A3A	2	7	1.249	-0.000	-0.000	-0.002	-0.004	-0.2806	-0.9712	1.01	0.05	0.96	0.2100	0.6974	
	8	8	0.0029	0.0001	0.0004	0.0004	0.7564	0.9712	0.0064	0.0002	0.0008	0.0365	0.7104		
	9	9	-0.0012	0.0002	-0.0002	-0.0002	-1.0546	0.9712	0.0044	-0.0000	0.0006	-0.0365	0.6616		
A3A	3	7	1.250	1.03	-0.00	-0.02	0.05	0.98	1.03	0.05	0.98	0.1489	0.6578		
	8	8	0.0029	0.0001	0.0004	0.7796	0.004	0.0004	0.0021	0.004	0.0018	0.0580	0.6616		
	9	9	-0.0011	0.0002	-0.0001	0.6114	0.0001	0.0001	0.0018	0.0001	0.0018	0.0580	0.6616		
A3A	4	7	1.249	3.13	-0.00	-0.01	0.05	1.01	1.06	0.05	1.01	0.1132	0.6654		
	8	8	0.0032	0.0001	0.0004	0.6209	0.0004	0.0008	0.0048	0.0044	0.0044	0.0909	0.6655		
	9	9	0.0014	0.0001	-0.0001	0.6405	0.0001	0.0006	0.00334	0.0006	0.0044	0.0909	0.6655		
A3A	5	7	1.250	6.29	-0.00	-0.02	0.05	1.05	1.11	0.05	1.05	0.1086	0.6377		
	8	8	0.0016	0.0001	0.0002	0.6806	0.0002	0.0013	0.0082	0.0081	0.0081	0.0951	0.6490		
	9	9	0.0014	0.0003	0.0001	0.5218	0.0001	0.0011	0.00628	0.0011	0.0062	0.0951	0.6490		
A3A	6	7	1.250	9.40	-0.00	-0.02	0.05	1.08	1.15	0.05	1.08	0.1015	0.6391		
	8	8	0.0014	0.0000	0.0002	0.8166	0.0002	0.0017	0.0110	0.0109	0.0109	0.0945	0.6349		
	9	9	-0.0004	0.0002	-0.0001	1.4413	0.0001	0.0016	0.0864	0.0016	0.0109	0.0945	0.6349		
A3A	7	7	1.251	12.65	-0.00	-0.02	0.05	1.11	1.18	0.05	1.11	0.0935	0.6221		
	8	8	0.0011	0.0000	0.0001	0.6178	0.0001	0.0019	0.0131	0.0130	0.0130	0.0860	0.6133		
	9	9	0.0002	0.0002	-0.0001	3.4768	0.0001	0.0018	0.1061	0.0018	0.0130	0.0860	0.6133		
A3A	8	7	1.250	-0.03	-0.00	-0.02	0.05	0.97	1.01	0.05	0.97	0.1663	0.6658		
	8	8	0.0028	0.0001	0.0004	0.7900	0.0004	0.0002	0.0008	0.0008	0.0008	0.0600	0.6658		
	9	9	-0.0015	0.0002	-0.0001	0.6183	0.0001	-0.0000	0.0040	-0.0000	0.0005	-0.0600	0.6658		
A3A	1	7	1.250	-3.09	-0.00	-0.02	0.05	3.03	2.95	0.05	3.03	0.1020	0.7018		
	8	8	0.0021	0.0001	0.0003	0.7651	0.0003	-0.0001	0.0012	-0.0010	0.0010	0.3113	0.6801		
	9	9	-0.0016	0.0002	-0.0002	0.7565	0.0002	-0.0005	-0.0010	-0.0010	0.0010	0.3113	0.6801		
A3A	2	7	1.250	-0.03	-0.00	-0.02	0.05	3.08	3.00	0.05	3.08	0.1396	0.6654		
	8	8	0.0026	0.0001	0.0004	0.8449	0.0004	0.0004	0.0023	0.0023	0.0023	0.0752	0.6654		
	9	9	-0.0009	0.0002	-0.0002	1.3503	0.0002	0.0002	0.0021	0.0021	0.0021	0.0752	0.6654		
A3A	3	7	1.250	1.03	-0.00	-0.02	0.05	3.08	3.01	0.05	3.08	0.1176	0.6745		
	8	8	0.0030	0.0001	0.0004	0.7089	0.0004	0.0006	0.0038	0.0038	0.0038	0.0860	0.6745		
	9	9	-0.0008	0.0002	-0.0001	1.1131	0.0001	0.0004	0.0035	0.0035	0.0035	0.0860	0.6745		
A3A	4	7	1.251	3.1A	-0.00	-0.02	0.05	3.11	3.05	0.05	3.11	0.1072	0.6700		
	8	8	0.0024	0.0000	0.0004	0.8282	0.0004	0.0010	0.0065	0.0065	0.0065	0.0950	0.6700		
	9	9	-0.0004	0.0002	-0.0000	0.8558	0.0000	0.0008	0.0062	0.0062	0.0062	0.0950	0.6700		

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key											
A42	2	7	1.249	0.09	-0.00	-0.02	15.14	0.05	15.11	0.1001	0.6523		
		8	0.0025	-0.0000	0.0004	0.8904	0.0854	0.0017	0.0111	0.0903	0.6511		
		9	-0.0011	0.0002	-0.0001	-0.9336	0.0854	0.0015	0.0111				
A42	3	7	1.249	1.12	-0.00	-0.02	15.15	0.05	15.12	0.0977	0.6429		
		8	0.0024	-0.0000	0.0004	0.8402	0.0929	0.0018	0.0119	0.0887	0.6414		
		9	-0.0010	0.0002	-0.0001	-1.1463	0.0925	0.0016	0.0118				
A42	4	7	1.248	3.24	-0.00	-0.02	15.17	0.05	15.14	0.0925	0.6288		
		8	0.0025	-0.0000	0.0003	0.927	0.1065	0.0019	0.0134	0.0845	0.6241		
		9	0.0000	0.0002	-0.0000	-3.6166	0.1068	0.0018	0.0133				
A42	5	7	1.249	6.36	-0.00	-0.02	15.19	0.06	15.17	0.1016	0.6014		
		8	0.0014	-0.0000	0.0002	0.9613	0.1169	0.0023	0.0140	0.0996	0.5905		
		9	0.0007	0.0003	0.0001	0.9496	0.1188	0.0023	0.0140				
A42	6	7	1.250	9.48	-0.00	-0.02	15.20	0.05	15.17	0.0900	0.5946		
		8	0.0017	-0.0000	0.0002	0.7244	0.1290	0.0023	0.0153	0.0864	0.5844		
		9	-0.0010	0.0002	0.0000	-0.2494	0.1310	0.0022	0.0153				
A42	7	7	1.250	12.68	-0.00	-0.02	15.21	0.05	15.18	0.0801	0.5885		
		8	0.0009	-0.0000	0.0001	0.8846	0.1362	0.0021	0.0160	0.0777	0.5734		
		9	-0.0011	0.0002	-0.0000	0.3071	0.1365	0.0021	0.0156				
A42	8	7	1.250	0.09	-0.00	-0.02	15.14	0.05	15.12	0.0999	0.6512		
		8	0.0031	-0.0001	0.0004	0.6786	0.0853	0.0015	0.0111	0.0899	0.6489		
		9	-0.0017	0.0002	-0.0000	0.2694	0.0851	0.0015	0.0110				
A43	3	7	1.250	-3.21	44.99	-3.07	-3.00	-3.04	-3.01	0.1021	0.6883		
		8	0.0401	-0.0011	-0.0053	0.6690	0.0392	0.0008	0.0054	0.1504	0.6929		
		9	-0.0384	-0.0007	-0.0050	0.6535	-0.0341	-0.0010	-0.0047				
A43	4	7	1.249	-0.09	44.99	-3.03	-2.96	-3.01	-2.99	0.1117	0.7075		
		8	0.0153	-0.0006	-0.0019	0.6202	0.0163	0.0003	0.0023	0.2524	0.7193		
		9	-0.0177	-0.0003	-0.0022	0.6428	-0.0149	-0.0007	-0.0021				
A43	5	7	1.250	0.98	44.99	-3.01	-2.95	-3.00	-2.99	0.1295	0.7615		
		8	0.0071	-0.0005	-0.0009	0.6338	0.0098	0.0002	0.0015	0.3495	0.7493		
		9	-0.0101	-0.0002	-0.0014	0.7135	-0.0089	-0.0006	-0.0013				
A43	6	7	1.249	3.07	44.99	-2.99	-2.93	-2.97	-2.97	0.6199	0.2989		
		8	0.0042	-0.0000	0.0006	0.7710	0.0004	0.0000	0.0000	-0.8216	-0.4662		
		9	0.0007	0.0003	0.0003	2.4084	0.0016	-0.0002	-0.0001				

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																			
843	7	1.250	6.23	44.99	0.1140	-2.95	-2.90	-2.92	-2.93	0.1450	0.6213	0.0232	0.0005	0.0030	0.2039	0.6550	0.0210	0.0005	0.0024	0.0605	0.6293
	8	0.0238	0.0009	0.0032	0.1273	0.6608	0.0445	0.0010	0.0026	0.1370	0.6439	0.0420	0.0014	0.0055	0.1553	0.6608	0.0445	0.0008	0.0049	0.0921	0.6476
	9	0.0456	0.0014	0.0060	0.1297	0.6611	0.0653	0.0012	0.0057	0.1296	0.6475	0.0456	0.0014	0.0085	0.1447	0.6611	0.0653	0.0012	0.0085	0.0938	0.6529
843	7	1.249	12.57	44.99	0.2099	-3.05	-2.96	-3.01	-3.00	0.1164	0.7171	1.249	0.0015	44.99	0.2111	0.6215	-0.0148	0.0007	-0.0021	0.2521	0.7218
	8	0.0583	0.0015	0.0077	0.1744	0.6361	0.0195	0.0004	0.0026	0.1064	0.6878	0.0583	0.0015	0.0025	0.1744	0.6361	0.0195	0.0004	0.0026	0.1064	0.6878
	9	0.0653	0.0018	0.0086	0.0845	0.6365	0.0165	0.0007	0.0022	0.2207	0.6880	0.0653	0.0018	0.0025	0.0845	0.6365	0.0165	0.0007	0.0022	0.2207	0.6880
844	1	1.250	-3.16	44.99	0.1620	-0.02	-0.07	0.04	0.03	0.3998	1.1593	1.250	0.0001	0.0002	-1.6206	3.4639	-0.0004	0.0002	-0.0001	-0.3998	1.1593
	2	0.0074	-0.0006	0.0025	-0.0162	0.7106	0.0043	0.0002	0.0005	0.2464	0.6680	0.0074	-0.0006	0.0025	-0.0162	0.7106	0.0043	0.0002	0.0005	0.2464	0.6680
	3	0.0049	0.0004	0.0006	-0.0421	0.6949	0.0053	-0.0000	0.0007	0.0831	0.7032	0.0049	0.0004	0.0006	-0.0421	0.6949	0.0053	-0.0000	0.0007	0.0831	0.7032
844	4	1.250	3.11	44.99	0.0941	0.672	0.03	0.10	0.01	0.1431	0.6533	1.250	0.0004	0.0028	0.0941	0.6672	0.0160	0.0002	0.0020	0.1431	0.6533
	5	0.0212	0.0007	0.0027	0.1768	0.6170	0.0183	0.0002	0.0024	0.0592	0.6566	0.0212	0.0007	0.0027	0.1768	0.6170	0.0183	0.0002	0.0024	0.0592	0.6566
	6	0.0446	0.0013	0.0060	0.1200	0.6743	0.0409	0.0007	0.0053	0.1222	0.6617	0.0446	0.0013	0.0060	0.1200	0.6743	0.0409	0.0007	0.0053	0.1222	0.6617
844	7	1.249	9.44	44.99	0.1200	0.6666	0.0549	0.017	0.05	0.1202	0.6559	1.249	0.0014	0.0077	0.1200	0.6666	0.0549	0.017	0.05	0.1202	0.6559
	8	0.0584	0.0017	0.0084	0.1381	0.6465	0.0633	0.0012	0.0082	0.0958	0.6525	0.0584	0.0017	0.0084	0.1381	0.6465	0.0633	0.0012	0.0082	0.0958	0.6525
	9	0.0649	0.0021	0.0105	0.1156	0.6386	0.0622	0.0015	0.0105	0.1172	0.6484	0.0649	0.0021	0.0105	0.1156	0.6386	0.0622	0.0015	0.0105	0.1172	0.6484

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																				
A44	8	1.249	-0.001	44.99	-0.9340	-0.02	-0.007	0.04	-0.001	-0.2123	2.3877	0.0010	-0.0002	0.0001	0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000	-0.0000
	6	0.0014	0.0002	0.0001	-0.7815	1.1172	-0.0003	0.0000	0.0000	-2.6419	-0.0106	-0.0014	0.0002	0.0001	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
	5	-0.0014	-0.0002	-0.0001	-0.7815	0.3989	0.0005	-0.0003	-0.0003	-2.6419	-0.0106	-0.0014	0.0002	0.0001	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000	0.0000
A45	1	1.250	-3.16	44.99	0.1924	0.5897	0.99	0.98	0.92	0.0939	0.7062	-0.0131	-0.0005	-0.0015	-0.0017	-0.0015	-0.0015	-0.0015	-0.0015	-0.0015	-0.0015	-0.0015
	6	0.0144	-0.0001	-0.0018	0.0535	0.6124	-0.0110	-0.0006	-0.0006	0.2726	0.6878	-0.0144	-0.0001	-0.0006	-0.0006	-0.0006	-0.0006	-0.0006	-0.0006	-0.0006	-0.0006	-0.0006
	5	-0.0144	-0.0001	-0.0018	0.0535	0.6124	-0.0110	-0.0006	-0.0006	0.2726	0.6878	-0.0144	-0.0001	-0.0006	-0.0006	-0.0006	-0.0006	-0.0006	-0.0006	-0.0006	-0.0006	-0.0006
A45	2	1.251	-0.01	44.99	0.0134	1.01	1.02	1.01	0.96	0.2646	0.6756	0.0060	0.0000	0.0002	0.0005	0.0006	0.0006	0.0006	0.0006	0.0006	0.0006	0.0006
	6	0.0060	0.0000	0.0004	0.5466	0.7874	0.0048	0.0001	0.0005	-0.1136	0.6971	0.0060	0.0000	0.0002	0.0005	0.0006	0.0006	0.0006	0.0006	0.0006	0.0006	0.0006
	5	0.0033	0.0003	0.0004	0.5466	0.6467	0.0048	-0.0001	0.0006	-0.1136	0.6971	0.0033	0.0003	0.0004	0.0006	0.0006	0.0006	0.0006	0.0006	0.0006	0.0006	0.0006
A45	3	1.251	1.03	44.99	0.0567	1.02	1.03	1.03	0.97	0.1696	0.6500	0.0131	0.0001	0.0003	0.0013	0.0015	0.0015	0.0015	0.0015	0.0015	0.0015	0.0015
	6	0.0131	0.0001	0.0014	0.2897	0.6678	0.0111	0.0000	0.0003	0.0264	0.6729	0.0131	0.0001	0.0003	0.0013	0.0015	0.0015	0.0015	0.0015	0.0015	0.0015	0.0015
	5	0.0103	0.0005	0.0014	0.2897	0.6906	0.0111	0.0000	0.0000	0.0264	0.6729	0.0103	0.0005	0.0014	0.0015	0.0015	0.0015	0.0015	0.0015	0.0015	0.0015	0.0015
B45	4	1.251	3.13	44.99	0.0991	1.05	1.06	1.06	0.99	0.1246	0.6606	0.0276	0.0005	0.0037	0.0029	0.0032	0.0032	0.0032	0.0032	0.0032	0.0032	0.0032
	6	0.0276	0.0005	0.0036	0.1764	0.750	0.0223	0.0005	0.0005	0.0779	0.6538	0.0276	0.0005	0.0037	0.0029	0.0032	0.0032	0.0032	0.0032	0.0032	0.0032	0.0032
	5	0.0273	0.0009	0.0036	0.1764	0.6770	0.0249	0.0003	0.0003	0.0779	0.6538	0.0273	0.0009	0.0036	0.0032	0.0032	0.0032	0.0032	0.0032	0.0032	0.0032	0.0032
A45	5	1.249	6.27	44.99	0.1167	1.08	1.09	1.10	1.03	0.1129	0.6597	0.0467	0.0010	0.0063	0.0057	0.0062	0.0062	0.0062	0.0062	0.0062	0.0062	0.0062
	6	0.0467	0.0010	0.0067	0.1475	0.755	0.0436	0.0009	0.0009	0.0944	0.6566	0.0467	0.0010	0.0063	0.0057	0.0062	0.0062	0.0062	0.0062	0.0062	0.0062	0.0062
	5	0.0510	0.0015	0.0067	0.1475	0.6608	0.0477	0.0009	0.0009	0.0944	0.6566	0.0510	0.0015	0.0067	0.0062	0.0062	0.0062	0.0062	0.0062	0.0062	0.0062	0.0062
A45	6	1.251	9.44	44.99	0.1145	1.12	1.12	1.14	1.05	0.1153	0.6546	0.0638	0.0014	0.0084	0.0079	0.0089	0.0089	0.0089	0.0089	0.0089	0.0089	0.0089
	6	0.0638	0.0014	0.0090	0.1384	0.6509	0.0694	0.0013	0.0013	0.0945	0.6467	0.0638	0.0014	0.0084	0.0079	0.0089	0.0089	0.0089	0.0089	0.0089	0.0089	0.0089
	5	0.0696	0.0019	0.0090	0.1384	0.6509	0.0694	0.0013	0.0013	0.0945	0.6467	0.0696	0.0019	0.0090	0.0089	0.0089	0.0089	0.0089	0.0089	0.0089	0.0089	0.0089
A45	7	1.251	12.61	44.99	0.1144	1.14	1.15	1.17	1.08	0.1115	0.6436	0.0779	0.0017	0.0109	0.0096	0.0111	0.0111	0.0111	0.0111	0.0111	0.0111	0.0111
	6	0.0779	0.0017	0.0109	0.1264	0.6237	0.0877	0.0016	0.0016	0.0915	0.6376	0.0779	0.0017	0.0109	0.0096	0.0111	0.0111	0.0111	0.0111	0.0111	0.0111	0.0111
	5	0.0881	0.0022	0.0109	0.1264	0.6237	0.0877	0.0016	0.0016	0.0915	0.6376	0.0881	0.0022	0.0109	0.0096	0.0111	0.0111	0.0111	0.0111	0.0111	0.0111	0.0111
A45	8	1.250	-0.03	44.99	-0.0199	1.01	1.02	1.01	0.96	-0.2180	0.6109	0.0061	-0.0000	0.0008	0.0005	0.0006	0.0006	0.0006	0.0006	0.0006	0.0006	0.0006
	6	0.0061	-0.0000	0.0005	-0.9301	0.7189	0.0043	0.0001	0.0005	-0.1107	0.6545	0.0061	-0.0000	0.0008	0.0005	0.0006	0.0006	0.0006	0.0006	0.0006	0.0006	0.0006
	5	0.0023	-0.0004	0.0005	-0.9301	1.0892	0.0047	-0.0001	0.0006	-0.1107	0.6545	0.0023	-0.0004	0.0005	0.0006	0.0006	0.0006	0.0006	0.0006	0.0006	0.0006	0.0006
A46	16	1.251	-3.14	44.99	0.2919	3.01	2.98	2.91	3.04	-0.4280	1.3817	0.0021	-0.0001	-0.0005	-0.0002	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001
	6	0.0021	-0.0001	-0.0005	-0.1639	0.7243	-0.0004	-0.0002	3.04	-3.4195	1.4483	0.0021	-0.0001	-0.0005	-0.0002	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001
	5	-0.0039	-0.0001	-0.0005	-0.1639	0.7243	-0.0004	-0.0002	3.04	-3.4195	1.4483	-0.0039	-0.0001	-0.0005	-0.0002	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001	-0.0001
A46	17	1.249	0.00	44.99	0.1128	3.04	3.01	2.95	3.07	0.1644	0.6674	0.0177	0.0003	0.0025	0.0020	0.0022	0.0022	0.0022	0.0022	0.0022	0.0022	0.0022
	6	0.0177	0.0003	0.0019	0.2421	0.7085	0.0155	0.0001	0.0005	0.0575	0.6661	0.0177	0.0003	0.0025	0.0020	0.0022	0.0022	0.0022	0.0022	0.0022	0.0022	0.0022
	5	0.0144	0.0007	0.0019	0.2421	0.6779	0.0169	0.0001	0.0005	0.0575	0.6661	0.0144	0.0007	0.0019	0.0020	0.0022	0.0022	0.0022	0.0022	0.0022	0.0022	0.0022

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key									
846	18	1.249	1.05	44.99	0.1037	3.05	3.02	2.97	3.09	0.1357	0.6690
		0.0254	0.0005	0.0035	0.1888	0.6939	0.0222	0.0006	0.0029	0.0735	0.6640
		0.0230	0.0008	0.0031	0.1888	0.6809	0.0244	0.0003	0.0032		
846	19	1.250	3.16	44.99	0.1111	3.07	3.04	3.00	3.11	0.1107	0.6703
		0.0395	0.0008	0.0054	0.1456	0.6868	0.0346	0.0007	0.0046	0.0883	0.6641
		0.0413	0.0012	0.0054	0.1456	0.6561	0.0388	0.0006	0.0051		
846	20	1.250	6.29	44.99	0.1141	3.11	3.08	3.04	3.13	0.1143	0.6645
		0.0575	0.0013	0.0077	0.1426	0.6754	0.0547	0.0012	0.0072	0.0958	0.6516
		0.0623	0.0017	0.0081	0.1426	0.6568	0.0610	0.0011	0.0079		
846	21	1.250	9.46	44.99	0.1098	3.14	3.11	3.07	3.17	0.1124	0.6505
		0.0738	0.0016	0.0096	0.1339	0.6564	0.0705	0.0015	0.0091	0.0937	0.6382
		0.0798	0.0021	0.0102	0.1339	0.6417	0.0810	0.0015	0.0103		
846	22	1.251	12.63	44.99	0.1070	3.16	3.12	3.10	3.18	0.1099	0.6383
		0.0869	0.0018	0.0111	0.1302	0.6410	0.0833	0.0018	0.0106	0.0511	0.6279
		0.0952	0.0024	0.0118	0.1302	0.6215	0.0980	0.0017	0.0123		
846	23	1.251	0.00	44.99	0.1089	3.04	3.00	2.95	3.08	0.1606	0.6598
		0.0177	0.0003	0.0024	0.2543	0.7025	0.0154	0.0004	0.0020	0.0559	0.6531
		0.0142	0.0007	0.0019	0.2543	0.6867	0.0167	0.0001	0.0021		
847	1	1.249	-3.10	44.99	0.1338	6.05	6.01	6.07	6.01	0.1922	0.7039
		0.0148	0.0003	0.0021	0.3356	0.7269	0.0144	0.0005	0.0020	0.0685	0.6772
		0.0109	0.0007	0.0016	0.3356	0.7338	0.0139	0.0001	0.0018		
847	2	1.250	0.04	44.99	0.1133	6.08	6.04	6.11	6.04	0.1170	0.6841
		0.0362	0.0008	0.0050	0.1517	0.7037	0.0340	0.0007	0.0046	0.0816	0.6675
		0.0355	0.0010	0.0048	0.1517	0.6755	0.0362	0.0005	0.0048		
847	3	1.250	1.10	44.99	0.0958	6.09	6.06	6.13	6.06	0.1096	0.6817
		0.0444	0.0008	0.0060	0.1442	0.6760	0.0404	0.0008	0.0055	0.0856	0.6701
		0.0440	0.0012	0.0060	0.1442	0.6847	0.0440	0.0007	0.0059		
847	4	1.251	3.20	44.99	0.1076	6.12	6.08	6.15	6.08	0.1051	0.6755
		0.0568	0.0012	0.0077	0.1409	0.6802	0.0526	0.0011	0.0071	0.0951	0.6627
		0.0599	0.0016	0.0079	0.1409	0.6627	0.0581	0.0011	0.0077		
847	5	1.250	6.35	44.99	0.1057	6.15	6.11	6.19	6.10	0.1086	0.6557
		0.0738	0.0015	0.0097	0.1326	0.6529	0.0706	0.0015	0.0092	0.0968	0.6455
		0.0796	0.0021	0.0101	0.1326	0.6343	0.0778	0.0015	0.0100		

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key											
847	6	7	1.251	9.52	44.99	0.1046	6.17	6.13	6.23	6.12	0.1062	0.6409		
		8	0.0879	0.0018	0.0113	0.1316	0.6435	0.0845	0.0017	0.0108	0.0937	0.6300		
		9	0.0941	0.0024	0.0116	0.1316	0.6168	0.0962	0.0018	0.0121				
847	7	7	1.252	12.65	44.99	0.1047	6.19	6.15	6.25	6.15	0.1032	0.6277		
		8	0.0979	0.0020	0.0122	0.1295	0.6247	0.0952	0.0019	0.0119	0.1093	0.5988		
		9	0.1055	0.0027	0.0125	0.1295	0.5947	0.1050	0.0022	0.0125				
847	8	7	1.252	0.07	44.99	0.1062	6.08	6.04	6.11	6.04	0.1136	0.6778		
		8	0.0367	0.0007	0.0050	0.1589	0.6943	0.0343	0.0007	0.0046	0.0806	0.6616		
		9	0.0353	0.0011	0.0044	0.1589	0.6853	0.0363	0.0005	0.0048				
848	1	7	1.250	-3.05	44.99	0.1197	9.06	9.11	9.04	9.03	0.1290	0.6991		
		8	0.0337	0.0008	0.0048	0.1854	0.7130	0.0340	0.0008	0.0047	0.0803	0.6778		
		9	0.0290	0.0010	0.0041	0.1854	0.7090	0.0317	0.0005	0.0043				
848	2	7	1.251	0.08	44.99	0.1069	9.10	9.13	9.09	9.06	0.1074	0.6806		
		8	0.0541	0.0011	0.0074	0.1361	0.6888	0.0526	0.0011	0.0071	0.0911	0.6694		
		9	0.0541	0.0014	0.0072	0.1361	0.6668	0.0546	0.0009	0.0073				
848	3	7	1.250	1.12	44.99	0.1080	9.11	9.15	9.10	9.07	0.1054	0.6750		
		8	0.0608	0.0013	0.0083	0.1402	0.6843	0.0580	0.0012	0.0078	0.0963	0.6671		
		9	0.0609	0.0017	0.0082	0.1402	0.6730	0.0615	0.0011	0.0082				
848	4	7	1.251	3.23	44.99	0.1051	9.13	9.17	9.13	9.09	0.1023	0.6647		
		8	0.0729	0.0015	0.0097	0.1333	0.6699	0.0692	0.0014	0.0092	0.0965	0.6515		
		9	0.0759	0.0020	0.0096	0.1333	0.6333	0.0746	0.0014	0.0097				
848	5	7	1.250	6.37	44.99	0.1033	9.16	9.19	9.16	9.11	0.1037	0.6460		
		8	0.0878	0.0018	0.0113	0.1475	0.6486	0.0850	0.0017	0.0109	0.0976	0.6264		
		9	0.0892	0.0026	0.0108	0.1475	0.6093	0.0923	0.0018	0.0115				
848	6	7	1.250	9.53	44.99	0.1026	9.18	9.21	9.18	9.13	0.0993	0.6333		
		8	0.0994	0.0020	0.0124	0.1336	0.6266	0.0874	0.0019	0.0123	0.1123	0.5944		
		9	0.1018	0.0027	0.0120	0.1336	0.5929	0.1019	0.0022	0.0121				
848	7	7	1.250	12.68	44.99	0.0982	9.19	9.22	9.20	9.14	0.0938	0.6233		
		8	0.1086	0.0021	0.0133	0.1367	0.6138	0.1073	0.0020	0.0133	0.1085	0.5798		
		9	0.1108	0.0030	0.0128	0.1367	0.5785	0.1110	0.0024	0.0128				
848	8	7	1.251	0.09	44.99	0.1073	9.10	9.13	9.09	9.06	0.1049	0.6759		
		8	0.0540	0.0011	0.0074	0.1355	0.6884	0.0528	0.0011	0.0071	0.0911	0.6670		
		9	0.0544	0.0014	0.0071	0.1355	0.6594	0.0547	0.0009	0.0073				

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																
A49	1	7	1.252	-2.99	44.99	0.1101	15.15	15.11	15.12	15.08	0.1114	0.6750	0.0688	0.0015	0.0088	0.0091	0.0894	0.6660	
			0.0645	0.0017	0.0084	0.1334	0.6582	0.0669	0.0011	0.0088	0.0091	0.0894	0.6660	0.0688	0.0015	0.0088	0.0091	0.0894	0.6660
			0.0645	0.0017	0.0084	0.1334	0.6582	0.0669	0.0011	0.0088	0.0091	0.0894	0.6660	0.0688	0.0015	0.0088	0.0091	0.0894	0.6660
A49	2	7	1.252	0.14	44.99	0.1022	15.17	15.13	15.15	15.10	0.1019	0.6575	0.0829	0.0016	0.0109	0.0109	0.0958	0.6575	
			0.0861	0.0017	0.0113	0.1420	0.6249	0.0856	0.0016	0.0109	0.0109	0.0958	0.6575	0.0829	0.0016	0.0109	0.0109	0.0958	0.6575
			0.0821	0.0023	0.0102	0.1420	0.6249	0.0856	0.0016	0.0109	0.0109	0.0958	0.6575	0.0829	0.0016	0.0109	0.0109	0.0958	0.6575
A49	3	7	1.253	1.20	44.99	0.0998	15.18	15.14	15.17	15.11	0.0990	0.6524	0.0876	0.0017	0.0114	0.0110	0.0990	0.6524	
			0.0914	0.0018	0.0118	0.1476	0.6096	0.0892	0.0019	0.0110	0.0110	0.0990	0.6524	0.0876	0.0017	0.0114	0.0110	0.0990	0.6524
			0.0870	0.0025	0.0106	0.1476	0.6096	0.0892	0.0019	0.0110	0.0110	0.0990	0.6524	0.0876	0.0017	0.0114	0.0110	0.0990	0.6524
A49	4	7	1.252	3.25	44.99	0.0980	15.20	15.15	15.18	15.12	0.0950	0.6444	0.0965	0.0018	0.0124	0.0115	0.0950	0.6444	
			0.1001	0.0019	0.0126	0.1427	0.5989	0.0957	0.0023	0.0115	0.0115	0.0950	0.6444	0.0965	0.0018	0.0124	0.0115	0.0950	0.6444
			0.0959	0.0027	0.0114	0.1427	0.5989	0.0957	0.0023	0.0115	0.0115	0.0950	0.6444	0.0965	0.0018	0.0124	0.0115	0.0950	0.6444
A49	5	7	1.252	6.39	44.99	0.0952	15.21	15.17	15.20	15.14	0.0898	0.6301	0.1095	0.0019	0.0138	0.0124	0.0898	0.6301	
			0.1119	0.0021	0.0138	0.1350	0.5785	0.1067	0.0023	0.0124	0.0124	0.0898	0.6301	0.1095	0.0019	0.0138	0.0124	0.0898	0.6301
			0.1072	0.0028	0.0124	0.1350	0.5785	0.1067	0.0023	0.0124	0.0124	0.0898	0.6301	0.1095	0.0019	0.0138	0.0124	0.0898	0.6301
A49	6	7	1.253	9.53	44.99	0.0981	15.22	15.18	15.20	15.14	0.0922	0.6117	0.1170	0.0021	0.0132	0.0132	0.0922	0.6117	
			0.1180	0.0023	0.0140	0.1223	0.5755	0.1164	0.0023	0.0132	0.0132	0.0922	0.6117	0.1170	0.0021	0.0132	0.0132	0.0922	0.6117
			0.1153	0.0028	0.0132	0.1223	0.5755	0.1164	0.0023	0.0132	0.0132	0.0922	0.6117	0.1170	0.0021	0.0132	0.0132	0.0922	0.6117
A49	7	7	1.252	12.65	44.99	0.0929	15.23	15.19	15.20	15.14	0.0923	0.5981	0.1215	0.0022	0.0134	0.0134	0.0923	0.5981	
			0.1243	0.0023	0.0147	0.1158	0.5663	0.1200	0.0023	0.0134	0.0134	0.0923	0.5981	0.1215	0.0022	0.0134	0.0134	0.0923	0.5981
			0.1172	0.0027	0.0132	0.1158	0.5663	0.1200	0.0023	0.0134	0.0134	0.0923	0.5981	0.1215	0.0022	0.0134	0.0134	0.0923	0.5981
A49	11	7	1.252	0.15	44.99	0.1042	15.18	15.14	15.16	15.11	0.1024	0.6570	0.0829	0.0016	0.0109	0.0109	0.1024	0.6570	
			0.0859	0.0017	0.0113	0.1380	0.6222	0.0856	0.0016	0.0109	0.0109	0.1024	0.6570	0.0829	0.0016	0.0109	0.0109	0.1024	0.6570
			0.0825	0.0022	0.0102	0.1380	0.6222	0.0856	0.0016	0.0109	0.0109	0.1024	0.6570	0.0829	0.0016	0.0109	0.0109	0.1024	0.6570
A50	1	7	1.251	6.35	44.99	0.1018	7.15	5.10	5.15	7.10	0.1098	0.6569	0.0654	0.0014	0.0105	0.0105	0.1098	0.6569	
			0.0788	0.0016	0.0103	0.1358	0.6441	0.0821	0.0015	0.0105	0.0105	0.1098	0.6569	0.0654	0.0014	0.0105	0.0105	0.1098	0.6569
			0.0747	0.0020	0.0096	0.1358	0.6441	0.0821	0.0015	0.0105	0.0105	0.1098	0.6569	0.0654	0.0014	0.0105	0.0105	0.1098	0.6569
A50	2	7	1.249	8.47	44.99	0.1009	7.17	5.11	5.17	7.11	0.1098	0.6468	0.0758	0.0017	0.0119	0.0119	0.1098	0.6468	
			0.0882	0.0017	0.0113	0.1293	0.6347	0.0947	0.0017	0.0119	0.0119	0.1098	0.6468	0.0758	0.0017	0.0119	0.0119	0.1098	0.6468
			0.0858	0.0022	0.0108	0.1293	0.6347	0.0947	0.0017	0.0119	0.0119	0.1098	0.6468	0.0758	0.0017	0.0119	0.0119	0.1098	0.6468
A51	1	7	1.249	6.35	44.99	0.0972	8.17	4.10	4.10	8.10	0.0972	0.6589	0.0604	0.0013	0.0110	0.0110	0.0972	0.6589	
			0.0841	0.0016	0.0109	0.1531	0.6835	0.0869	0.0016	0.0110	0.0110	0.0972	0.6589	0.0604	0.0013	0.0110	0.0110	0.0972	0.6589
			0.0680	0.0020	0.0093	0.1531	0.6835	0.0869	0.0016	0.0110	0.0110	0.0972	0.6589	0.0604	0.0013	0.0110	0.0110	0.0972	0.6589

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key											
851	2	1.251	6.35	44.99	0.0964	10.16	2.08	2.07	10.20	0.1147	0.6573		
	6	0.0926	0.0017	0.0118	0.1411	0.6422	0.0497	0.0011	0.0065	0.0988	0.6202		
	9	0.0612	0.0017	0.0079	0.0964	0.6463	0.0947	0.0018	0.0117	0.0988	0.6202		
A52	1	1.247	4.23	44.99	0.1067	8.14	6.11	6.18	8.07	0.1031	0.6647		
	6	0.0730	0.0015	0.0097	0.1344	0.6697	0.0593	0.0012	0.0078	0.0973	0.6506		
	9	0.0688	0.001A	0.0088	0.1344	0.6404	0.0747	0.0014	0.0097	0.0973	0.6506		
853	1	0.899	-0.04	-90.00	1.4605	-0.03	-0.07	0.06	-0.03	0.2288	0.1017		
	6	-0.0009	-0.0002	-0.0002	1.4605	1.0889	0.0023	0.0001	0.0000	-0.0196	0.5108		
	9	-0.0015	0.0001	-0.0003	-0.4434	1.0974	0.0044	-0.0000	0.0004	-0.0196	0.5108		
853	2	0.899	-0.04	-78.80	1.4240	-0.03	-0.06	0.05	-0.02	0.2595	0.1731		
	6	-0.0009	-0.0002	-0.0001	1.4240	0.9313	0.0025	0.0001	0.0000	-0.0042	0.5552		
	9	-0.0013	0.0001	-0.0003	-0.4462	1.4316	0.0044	-0.0000	0.0004	-0.0042	0.5552		
853	3	0.900	-0.04	-67.50	6.4459	-0.02	-0.06	0.05	-0.02	0.3249	0.2634		
	6	-0.0002	-0.0002	-0.0000	6.4459	2.4720	0.0025	0.0001	0.0001	0.3249	0.2634		
	9	-0.0009	0.0000	-0.0004	-0.3873	2.4141	0.0037	-0.0000	0.0004	-0.0387	0.5298		
853	4	0.900	-0.04	-56.30	-3.5378	-0.02	-0.06	0.05	-0.03	0.3908	0.3584		
	6	0.0003	-0.0002	0.0000	-3.5378	0.4749	0.0025	0.0001	0.0001	0.3908	0.3584		
	9	-0.0014	0.0000	-0.0004	-0.3458	1.4642	0.0029	-0.0000	0.0002	-0.0951	0.4815		
853	5	0.899	-0.04	-45.00	-0.6368	-0.02	-0.06	0.05	-0.02	0.3951	0.3280		
	6	0.0008	-0.0001	0.0002	-0.6368	1.5461	0.0023	0.0001	0.0001	0.3951	0.3280		
	9	-0.0012	0.0001	-0.0003	-0.4279	1.5182	0.0018	-0.0000	0.0001	-0.1633	0.3805		
853	6	0.900	-0.04	-33.80	-0.4627	-0.02	-0.07	0.06	-0.02	0.3976	0.3154		
	6	0.0028	-0.0002	0.0002	-0.4627	0.4860	0.0022	0.0001	0.0001	0.3976	0.3154		
	9	-0.0021	0.0001	-0.0002	-0.4006	0.6938	0.0006	-0.0000	-0.0000	-0.7503	-0.1685		
853	7	0.898	-0.04	-22.50	-0.1353	-0.02	-0.06	0.05	-0.02	0.4535	0.3532		
	6	0.0028	-0.0000	0.0005	-0.1353	0.9093	0.0021	0.0001	0.0001	0.4535	0.3532		
	9	-0.0015	0.0001	-0.0002	-0.5429	0.9449	0.0000	-0.0000	-0.0000	-7.3468	-4.6806		
A53	8	0.899	-0.04	-11.30	-0.2338	-0.02	-0.07	0.06	-0.02	0.5291	0.3802		
	6	0.0044	-0.0002	0.0005	-0.2338	0.5731	0.0018	0.0001	0.0001	0.5291	0.3802		
	9	-0.0022	0.0002	-0.0002	-0.5149	0.4704	-0.0003	-0.0001	-0.0001	1.4211	1.5926		
A53	9	0.898	-0.04	-0.00	-0.0778	-0.02	-0.06	0.05	-0.03	0.5961	0.4496		
	6	0.0038	-0.0000	0.0006	-0.0778	0.8055	0.0015	0.0001	0.0001	0.5961	0.4496		
	9	-0.0012	0.0001	-0.0002	-0.7799	1.0063	-0.0004	-0.0001	-0.0001	1.3187	1.2829		

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key															
A53	10	7	0.900	-0.004	11.29	-0.0380	-0.02	-0.06	0.06	-0.03	0.5917	0.3673						
		8	0.0034	-0.0000	0.0006	-0.0001	0.9349	0.0013	0.0001	0.0001	1.6845	1.4427						
		9	-0.0011	0.0002	-0.0001	-0.0001	0.8762	-0.0003	-0.0001	-0.0000								
A53	11	7	0.900	-0.004	22.49	-0.1986	-0.02	-0.06	0.06	-0.02	0.6876	0.2195						
		8	0.0036	-0.0001	0.0005	-0.0001	0.7417	0.0010	0.0001	0.0000	4.3415	3.4119						
		9	-0.0010	0.0002	-0.0001	-0.0001	0.6788	-0.0001	-0.0001	-0.0000								
A53	12	7	0.899	-0.004	33.79	-0.2923	-0.02	-0.07	0.05	-0.03	0.9105	-0.4208						
		8	0.0025	-0.0001	0.0004	-0.0002	0.9233	0.0005	0.0000	-0.0000	2.1711	1.7896						
		9	-0.0009	0.0002	-0.0002	-0.0002	1.0872	-0.0002	-0.0000	-0.0000								
A53	13	7	0.899	-0.004	44.99	-0.5628	-0.02	-0.07	0.05	-0.02	-0.4298	1.9185						
		8	0.0015	-0.0001	0.0003	-0.0001	1.2009	-0.0004	0.0000	-0.0001	-1.5538	2.1145						
		9	-0.0013	0.0002	-0.0001	-0.0001	0.6199	-0.0003	-0.0001	-0.0001								
A53	14	7	0.899	-0.004	56.29	-1.0222	-0.02	-0.07	0.05	-0.03	0.0949	1.3624						
		8	0.0001	-0.0000	0.0003	-0.0002	-0.0535	-0.0010	-0.0000	-0.0002	1.3290	1.6980						
		9	-0.0012	0.0002	-0.0002	-0.0002	0.8435	-0.0004	-0.0001	-0.0001								
A53	15	7	0.900	-0.004	67.49	-0.7079	-0.02	-0.06	0.05	-0.03	0.1766	1.1888						
		8	0.0006	-0.0000	0.0002	-0.0002	-1.7625	-0.0016	-0.0000	-0.0003	0.8372	1.2494						
		9	-0.0011	0.0002	-0.0002	-0.0002	0.9944	-0.0006	-0.0001	-0.0001								
A53	16	7	0.899	-0.004	78.79	-5.6536	-0.02	-0.08	0.06	-0.03	0.2393	1.1218						
		8	0.0002	-0.0002	0.0001	-0.0002	-3.5886	-0.0020	-0.0001	-0.0004	0.7517	1.2498						
		9	-0.0014	0.0002	-0.0002	-0.0002	0.6936	-0.0010	-0.0001	-0.0002								
A53	17	7	0.900	-0.004	89.99	-0.5762	-0.02	-0.07	0.06	-0.03	0.2942	1.1742						
		8	0.0009	-0.0001	0.0002	-0.0001	-1.1668	-0.0022	-0.0001	-0.0005	0.8814	1.4191						
		9	-0.0016	0.0002	-0.0001	-0.0001	0.6114	-0.0010	-0.0001	-0.0002								
A54	3	7	1.250	-3.13	-0.00	-0.2217	-0.02	-0.11	0.05	-0.08	0.1131	0.6781						
		8	0.0024	-0.0001	0.0004	-0.0003	0.8664	-0.0266	-0.0006	-0.0036	0.1671	0.6704						
		9	-0.0006	0.0001	-0.0003	-0.0003	2.5864	-0.0265	-0.0008	-0.0035								
A54	4	7	1.250	-0.02	-0.00	-0.1096	-0.02	-0.06	0.06	-0.03	0.5505	0.5804						
		8	0.0036	-0.0000	0.0005	-0.0002	0.8095	0.0008	0.0000	0.0001	0.5922	0.8290						
		9	-0.0006	0.0002	-0.0002	-0.0002	1.7946	0.0001	-0.0003	0.0000	-11.5922							
A54	5	7	1.249	-1.02	-0.00	-0.1169	-0.02	-0.04	0.06	-0.01	0.1577	0.6510						
		8	0.0040	-0.0000	0.0005	-0.0001	0.7113	0.0103	0.0003	0.0013	0.0057	0.6990						
		9	-0.0009	0.0003	-0.0001	-0.0001	0.6353	0.0075	0.0000	0.0010								

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key													
A54	6	7	1.249	3.13	0.000	0.000	0.000	-0.0917	-0.02	-0.00	0.06	-0.0037	0.1254	0.6493		
		8	0.0031	-0.0000	0.0004	0.0004	-0.0917	0.6575	0.0285	0.0007	0.0037	0.0034	0.0912	0.6581		
		9	0.0001	0.0003	-0.0000	-0.0000	9.1794	-1.4380	0.0264	0.0004	0.0034	0.0034	0.0912	0.6581		
A54	7	7	1.250	6.27	0.000	0.000	-0.2508	-0.02	0.03	0.06	0.05	0.1146	0.6515			
		8	0.001A	-0.0000	0.0003	0.0003	-0.2508	0.8527	0.0555	0.0012	0.0072	0.0072	0.1059	0.6521		
		9	0.0020	0.0003	0.0000	0.0000	0.8543	0.0770	0.0555	0.0011	0.0072	0.0072	0.1059	0.6521		
A54	8	7	1.249	9.38	0.000	0.000	-0.2346	-0.02	0.07	0.06	0.07	0.1075	0.6337			
		8	0.0019	-0.0000	0.0002	0.0002	-0.2346	0.6227	0.0794	0.0017	0.0100	0.0100	0.0989	0.6324		
		9	0.0003	0.0003	-0.0000	-0.0000	4.7537	-0.4591	0.0791	0.0015	0.0100	0.0100	0.0989	0.6324		
A54	9	7	1.247	12.57	0.000	0.000	-0.3977	-0.02	0.10	0.06	0.10	0.0948	0.6198			
		8	0.0011	-0.0000	0.0001	0.0001	-0.3977	0.6831	0.1003	0.0019	0.0124	0.0124	0.0907	0.6178		
		9	0.0001	0.0002	-0.0001	-0.0001	10.3371	-4.3129	0.1002	0.0018	0.0123	0.0123	0.0907	0.6178		
A55	3	7	0.896	-3.08	0.000	0.000	-0.1419	-0.02	-0.10	0.04	-0.08	0.1935	0.6478			
		8	0.0024	-0.0000	0.0004	0.0004	-0.1419	1.0280	-0.0280	-0.0010	-0.0036	0.2594	0.6357			
		9	-0.0014	0.0000	-0.0002	-0.0002	-0.2974	0.9600	-0.0264	-0.0013	-0.0033	0.2594	0.6357			
A55	4	7	0.897	-0.03	0.000	0.000	0.1272	-0.03	-0.06	0.05	-0.04	0.9139	0.5374			
		8	0.002A	0.0000	0.0007	0.0007	0.1272	1.2718	0.0014	0.0002	0.0001	0.6853	-1.1731			
		9	-0.0002	0.0000	-0.0002	-0.0002	-1.1028	4.7650	-0.0002	-0.0000	0.0000	0.6853	-1.1731			
A55	5	7	0.897	0.98	0.000	0.000	-0.0838	-0.02	-0.05	0.05	-0.01	0.3364	0.7021			
		8	0.0041	-0.0000	0.0006	0.0006	-0.0838	0.8241	0.0099	0.0006	0.0013	0.3364	0.7021			
		9	-0.0012	0.0001	-0.0001	-0.0001	-0.4239	0.6225	0.0078	0.0002	0.0012	0.1899	0.7823			
A55	6	7	0.897	3.09	0.000	0.000	-0.1377	-0.02	-0.01	0.05	0.00	0.2377	0.6587			
		8	0.0032	-0.0000	0.0005	0.0005	-0.1377	0.8090	0.0316	0.0015	0.0041	0.1955	0.6706			
		9	-0.0011	0.0001	-0.0000	-0.0000	-0.6515	0.3275	0.0292	0.0011	0.0039	0.1955	0.6706			
A55	7	7	0.900	6.16	0.000	0.000	2.3009	-0.02	0.02	0.05	0.04	0.2236	0.6127			
		8	0.0001	0.0000	0.0000	0.0000	2.3009	2.8473	0.0580	0.0025	0.0071	0.2236	0.6127			
		9	0.0001	0.0001	-0.0000	-0.0000	5.3734	-0.4544	0.0594	0.0023	0.0074	0.1980	0.6257			
A55	8	7	0.900	9.27	0.000	0.000	-0.3631	-0.02	0.05	0.05	0.06	0.1937	0.5812			
		8	0.000A	-0.0000	0.0001	0.0001	-0.3631	1.1339	0.0800	0.0031	0.0093	0.1833	0.5812			
		9	-0.0012	0.0001	0.0000	0.0000	-0.6761	-0.2389	0.0816	0.0029	0.0095	0.1833	0.5812			
A55	9	7	0.897	12.43	0.000	0.000	-0.6790	-0.02	0.06	0.05	0.08	0.1592	0.5679			
		8	0.0009	-0.0001	0.0002	0.0002	-0.6790	1.4181	0.0984	0.0031	0.0111	0.1536	0.5679			
		9	-0.0015	0.0001	-0.0000	-0.0000	-0.3346	0.3070	0.1004	0.0030	0.0113	0.1536	0.5679			

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key													
855	10	7	0.897	-0.001	-0.000	-0.006	-0.1168	0.7633	-0.02	-0.06	0.04	-0.03	0.7385	0.4666	
		8	0.0041	-0.0000	0.0000	0.0006	-0.5715	0.6387	0.0016	0.0002	0.0001	0.0000	0.8588	0.0518	
		9	-0.0012	0.0001	-0.0001	-0.0001	-0.0003	-0.0003	-0.0003	-0.0000	-0.0000	-0.0000			
856	1	7	0.899	-3.05	-0.001	-0.004	-0.2496	-0.9085	6.02	0.05	6.00	0.3264	0.7369		
		8	0.0025	-0.0001	0.0001	0.0004	-0.4578	0.3787	0.0088	0.0005	0.0013	0.2698	0.7658		
		9	-0.0025	0.0002	-0.0001	-0.0001	0.0075	0.0075	0.0075	0.0004	0.0011				
856	2	7	0.898	-0.01	-0.001	-0.002	-0.0599	-0.9588	6.07	0.04	6.04	0.2225	0.6568		
		8	0.0034	-0.0000	0.0006	-0.0002	-0.6920	1.2507	0.412	0.0018	0.0054	0.1943	0.6676		
		9	-0.0008	0.0001	-0.0001	-0.0001	0.0389	0.0389	0.0389	0.0015	0.0052				
856	3	7	0.897	1.02	-0.001	-0.006	-0.0564	-0.9111	6.08	0.04	6.06	0.2279	0.6298		
		8	0.0033	-0.0000	0.0006	-0.0001	-0.5779	0.4914	0.495	0.0022	0.0062	0.1974	0.6462		
		9	-0.0012	0.0001	-0.0001	-0.0001	0.0483	0.0483	0.0483	0.0019	0.0062				
856	4	7	0.898	3.13	-0.001	-0.003	0.1071	-0.9839	6.10	0.05	6.08	0.2170	0.5970		
		8	0.0016	0.0000	0.0003	-0.0003	-1.4526	1.2066	0.645	0.0028	0.0077	0.2043	0.6032		
		9	-0.0005	0.0001	-0.0001	-0.0001	-1.6371	-0.6772	0.0638	0.0026	0.0076				
856	5	7	0.897	6.19	0.001	0.001	-0.0504	-0.9583	6.12	0.05	6.10	0.1890	0.5758		
		8	0.0006	-0.0002	0.0001	-0.0001	-1.6371	1.2583	0.0836	0.0031	0.0095	0.1855	0.5681		
		9	-0.0008	0.0002	0.0001	-0.0001	-1.6371	-0.6772	0.0837	0.0031	0.0095				
856	6	7	0.897	9.26	-0.001	-0.003	-0.6399	-1.9368	6.14	0.05	6.11	0.1509	0.5668		
		8	0.0007	-0.0001	0.0003	-0.0001	-45.3813	50.3626	0.979	0.0029	0.0111	0.1483	0.5581		
		9	-0.0000	0.0000	-0.0001	-0.0001	-45.3813	50.3626	0.0998	0.0029	0.0111				
856	7	7	0.897	12.41	-0.001	-0.001	-1.3579	-1.6314	6.14	0.04	6.11	0.1157	0.5684		
		8	0.0006	-0.0001	0.0001	-0.0001	-0.4023	0.6790	0.1073	0.0024	0.0123	0.1202	0.5564		
		9	-0.0011	0.0000	-0.0001	-0.0001	-0.6790	0.6790	0.1110	0.0026	0.0123				
856	8	7	0.897	-0.01	-0.001	-0.001	-0.2101	-0.6912	6.06	0.05	6.04	0.2309	0.6518		
		8	0.0042	-0.0001	0.0005	-0.0001	-0.5637	0.4866	0.398	0.0018	0.0050	0.2068	0.6704		
		9	-0.0014	0.0001	-0.0001	-0.0001	-0.5637	0.4866	0.0376	0.0015	0.0050				
857	1	7	0.895	-2.99	-0.001	-0.001	-0.2789	-0.9910	15.09	0.05	15.08	0.2240	0.6124		
		8	0.0023	-0.0001	0.0005	-0.0002	-0.4769	0.8789	0.0611	0.0027	0.0077	0.2072	0.6215		
		9	-0.0015	0.0001	-0.0001	-0.0001	-0.4769	0.8789	0.0622	0.0025	0.0077				
857	2	7	0.898	0.02	-0.001	-0.001	-0.1845	-0.6439	15.12	0.05	15.10	0.2000	0.5756		
		8	0.0030	-0.0001	0.0003	-0.0001	-0.7174	0.8519	0.0801	0.0032	0.0092	0.1905	0.5787		
		9	-0.0009	0.0001	-0.0001	-0.0001	-0.7174	0.8519	0.0804	0.0030	0.0092				

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																				
858	6	0.896	9.35	44.99	0.1191	15.20	15.16	15.14	15.10	0.1118	0.5872	0.1289	0.0030	0.0150	0.0115	0.1474	0.5843	0.1304	0.0029	0.0153	0.1297	0.5632
	6	0.1289	0.0030	0.0115	0.1474	0.5843	0.1304	0.0029	0.0153	0.1297	0.5632	0.1019	0.0030	0.0115	0.0150	0.1474	0.5843	0.1304	0.0029	0.0153	0.1297	0.5632
858	7	0.896	12.45	44.99	0.1103	15.21	15.17	15.14	15.11	0.1036	0.5865	0.1407	0.0031	0.0165	0.0117	0.1357	0.5863	0.1426	0.0029	0.0167	0.1265	0.5548
	8	0.1407	0.0031	0.0117	0.1357	0.5863	0.1426	0.0029	0.0167	0.1265	0.5548	0.1056	0.0028	0.0117	0.0165	0.1357	0.5863	0.1426	0.0029	0.0167	0.1265	0.5548
858	8	0.898	0.09	44.99	0.1829	15.16	15.12	15.13	15.10	0.1898	0.5831	0.0847	0.0031	0.0101	0.0091	0.2197	0.5672	0.0809	0.0030	0.0093	0.1900	0.5767
	5	0.0847	0.0035	0.0091	0.2197	0.5672	0.0809	0.0030	0.0093	0.1900	0.5767	0.0808	0.0035	0.0101	0.0091	0.2197	0.5672	0.0809	0.0030	0.0093	0.1900	0.5767
859	1	0.897	-3.03	44.99	0.2631	6.05	6.01	6.06	6.01	0.2671	0.6818	0.0169	0.0008	0.0024	0.0016	0.4316	0.7252	0.0165	0.0008	0.0022	0.1704	0.7096
	8	0.0169	0.0008	0.0024	0.4316	0.7252	0.0165	0.0008	0.0022	0.1704	0.7096	0.012A	0.0011	0.0016	0.0016	0.4316	0.7252	0.0165	0.0008	0.0022	0.1704	0.7096
859	2	0.897	0.05	44.99	0.2033	6.08	6.05	6.10	6.04	0.2014	0.6568	0.0410	0.0016	0.0055	0.0045	0.2877	0.6332	0.0396	0.0015	0.0049	0.2149	0.6640
	9	0.035A	0.0020	0.0045	0.2877	0.6332	0.0396	0.0015	0.0049	0.2149	0.6640	0.0410	0.0016	0.0055	0.0045	0.2877	0.6332	0.0396	0.0015	0.0049	0.2149	0.6640
859	3	0.898	1.07	44.99	0.1775	6.09	6.06	6.11	6.05	0.1831	0.6486	0.0501	0.0017	0.0065	0.0053	0.3029	0.6574	0.0468	0.0017	0.0060	0.2176	0.6512
	5	0.0501	0.0017	0.0065	0.3029	0.6574	0.0468	0.0017	0.0060	0.2176	0.6512	0.0409	0.0024	0.0053	0.0053	0.3029	0.6574	0.0468	0.0017	0.0060	0.2176	0.6512
859	4	0.897	3.14	44.99	0.1731	6.11	6.07	6.13	6.06	0.1710	0.6180	0.0621	0.0021	0.0077	0.0066	0.2952	0.6204	0.0584	0.0019	0.0067	0.2213	0.6215
	9	0.0621	0.0021	0.0077	0.2952	0.6204	0.0584	0.0019	0.0067	0.2213	0.6215	0.0535	0.0031	0.0066	0.0066	0.2952	0.6204	0.0584	0.0019	0.0067	0.2213	0.6215
859	5	0.898	6.22	44.99	0.1517	6.13	6.09	6.15	6.08	0.1513	0.6034	0.0800	0.0024	0.0097	0.0080	0.2681	0.6097	0.0775	0.0023	0.0078	0.2293	0.5768
	9	0.0800	0.0024	0.0080	0.2681	0.6097	0.0775	0.0023	0.0078	0.2293	0.5768	0.0694	0.0037	0.0080	0.0080	0.2681	0.6097	0.0775	0.0023	0.0078	0.2293	0.5768
859	6	0.901	9.30	44.99	0.1360	6.15	6.11	6.16	6.09	0.1369	0.5991	0.0971	0.0026	0.0118	0.0087	0.2368	0.6091	0.0947	0.0025	0.0090	0.2051	0.5571
	9	0.0794	0.0037	0.0087	0.2368	0.5503	0.0812	0.0033	0.0090	0.2051	0.5571	0.0794	0.0037	0.0087	0.0087	0.2368	0.6091	0.0947	0.0025	0.0090	0.2051	0.5571
859	7	0.897	12.40	44.99	0.1212	6.17	6.13	6.16	6.09	0.1219	0.5999	0.1141	0.0027	0.0136	0.0097	0.1849	0.5961	0.1125	0.0027	0.0099	0.1707	0.5533
	9	0.1141	0.0027	0.0097	0.1849	0.5619	0.0903	0.0030	0.0099	0.1707	0.5533	0.0865	0.0031	0.0097	0.0097	0.1849	0.5961	0.1125	0.0027	0.0099	0.1707	0.5533
859	8	0.899	0.04	44.99	0.2176	6.08	6.04	6.09	6.03	0.1962	0.6533	0.0401	0.0017	0.0055	0.0044	0.2832	0.6959	0.0395	0.0015	0.0049	0.2150	0.6606
	5	0.0360	0.0020	0.0044	0.2832	0.6212	0.0371	0.0015	0.0049	0.2150	0.6606	0.0401	0.0017	0.0055	0.0044	0.2832	0.6959	0.0395	0.0015	0.0049	0.2150	0.6606

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key									
A60	1	0.890	-3.10	44.99	0.3097	-0.06	-0.09	0.02	-0.06	0.2440	0.6927
	8	-0.0179	-0.0011	-0.0021	0.1262	0.5995	-0.0187	-0.0009	-0.0025	0.2471	0.6765
	9	-0.0231	-0.0005	-0.0026	0.1262	0.5647	-0.0185	-0.0009	-0.0025	0.2471	0.6765
A60	2	0.897	-0.01	44.99	-0.5335	-0.06	-0.06	0.05	-0.03	0.9623	-6.3254
	8	0.0022	-0.0002	0.0003	-1.2352	0.6745	0.0000	0.0000	-0.0001	0.7367	1.0511
	9	-0.0010	0.0002	-0.0000	-1.2352	0.4520	-0.0005	-0.0000	-0.0001	0.7367	1.0511
A60	3	0.897	1.02	44.99	0.0712	-0.01	-0.05	0.06	-0.02	0.3619	0.5511
	8	0.0090	0.0001	0.0012	0.5532	0.6862	0.0049	0.0003	0.0005	0.1249	0.7072
	9	0.0053	0.0005	0.0005	0.5532	0.5390	0.0059	0.0001	0.0008	0.1249	0.7072
A60	4	0.900	3.08	44.99	0.1769	0.01	-0.03	0.09	-0.01	0.1687	0.6516
	8	0.0241	0.0008	0.0032	0.3624	0.6702	0.0194	0.0006	0.0025	0.1933	0.5706
	9	0.0193	0.0014	0.0024	0.3624	0.6448	0.0183	0.0007	0.0024	0.1933	0.5706
A60	5	0.898	6.18	44.99	0.1510	0.04	-0.00	0.12	0.01	0.1417	0.6421
	8	0.0473	0.0014	0.0060	0.3042	0.6420	0.0440	0.0012	0.0056	0.2332	0.6370
	9	0.0407	0.0024	0.0051	0.3042	0.6328	0.0389	0.0018	0.0049	0.2332	0.6370
A60	6	0.898	9.28	44.99	0.1500	0.07	0.02	0.16	0.05	0.1449	0.6158
	8	0.0677	0.0020	0.0087	0.2692	0.6486	0.0641	0.0018	0.0079	0.2415	0.5971
	9	0.0592	0.0031	0.0065	0.2692	0.5524	0.0571	0.0027	0.0068	0.2415	0.5971
A60	7	0.898	12.38	44.99	0.1304	0.09	0.04	0.18	0.06	0.1288	0.6122
	8	0.0864	0.0022	0.0106	0.2575	0.6141	0.0837	0.0021	0.0102	0.2286	0.5695
	9	0.0715	0.0036	0.0080	0.2575	0.5610	0.0716	0.0032	0.0081	0.2286	0.5695
A60	8	0.897	0.01	44.99	-0.4274	-0.04	-0.06	0.04	-0.04	0.0265	-1.8403
	8	0.0015	-0.0001	0.0003	-1.0864	1.2340	0.0003	0.0000	-0.0001	1.7127	-2.3374
	9	-0.0013	0.0002	-0.0000	-1.0864	0.3517	-0.0002	-0.0001	-0.0001	1.7127	-2.3374
A61	6	0.600	-3.08	-0.003	0.4840	-0.01	-0.11	0.05	-0.08	0.2663	0.6167
	8	0.0008	0.0000	0.0003	-0.0604	2.2052	-0.0232	-0.0012	-0.0028	0.3353	0.6096
	9	-0.0016	0.0000	-0.0002	-0.0604	0.6733	-0.0215	-0.0014	-0.0026	0.3353	0.6096
A61	7	0.600	-0.03	0.00	0.4391	-0.02	-0.05	0.06	-0.04	0.6064	0.4694
	8	0.0018	0.0001	0.0005	-0.7077	1.4317	0.0020	0.0002	0.0001	-0.3721	1.2696
	9	0.0007	-0.0001	-0.0004	-0.7077	-2.8646	0.0008	-0.0000	0.0002	-0.3721	1.2696
A61	8	0.600	1.01	-0.00	-0.0365	-0.01	-0.04	0.06	-0.02	0.3658	0.6340
	8	0.0031	-0.0000	0.0004	-0.2400	0.6764	0.0095	0.0007	0.0012	0.2010	0.6647
	9	-0.0022	0.0001	-0.0001	-0.2400	0.2498	0.0096	0.0003	0.0011	0.2010	0.6647

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Pd	Cd	See Key															
A63	10	7 8 9	0.559 0.0761 0.0734	0.041 0.041 0.041	44.99 0.086 0.082	0.2718 0.2603	15.18 0.702 0.5622	15.14 0.753 0.759	15.15 0.040 0.038	15.14 0.085 0.085	0.2683 0.2533	0.5662 0.5657						
A63	11	7 8 9	0.600 0.0796 0.0761	1.11 0.041 0.041	44.99 0.088 0.083	0.2611 0.2749	15.19 0.5574 0.5498	15.15 0.784 0.783	15.15 0.041 0.039	15.13 0.087 0.087	0.2631 0.2497	0.5581 0.5568						
A63	12	7 8 9	0.600 0.0848 0.0818	3.16 0.039 0.041	44.99 0.093 0.088	0.2312 0.2536	15.18 0.505 0.5411	15.15 0.836 0.839	15.16 0.040 0.039	15.14 0.091 0.091	0.2419 0.2339	0.5457 0.5438						
A63	13	7 8 9	0.601 0.0906 0.0866	6.24 0.033 0.034	44.99 0.101 0.094	0.1860 0.2004	15.18 0.584 0.5439	15.15 0.889 0.892	15.15 0.033 0.034	15.13 0.097 0.096	0.1882 0.1953	0.5457 0.5397						
A64	1	7 8 9	0.600 0.0597 0.0594	-2.99 0.035 0.034	44.99 0.072 0.068	0.2994 0.2922	15.15 0.671 0.5766	15.11 0.592 0.618	15.12 0.034 0.031	15.11 0.070 0.073	0.2878 0.2524	0.5923 0.5929						
A64	2	7 8 9	0.600 0.0784 0.0726	0.10 0.040 0.041	44.99 0.086 0.083	0.2648 0.2877	15.18 0.533 0.5725	15.14 0.751 0.757	15.15 0.038 0.038	15.14 0.085 0.085	0.2690 0.2533	0.5673 0.5657						
A64	3	7 8 9	0.601 0.0793 0.0759	1.11 0.041 0.041	44.99 0.088 0.083	0.2615 0.2762	15.18 0.584 0.5514	15.15 0.783 0.781	15.16 0.041 0.039	15.13 0.087 0.087	0.2637 0.2505	0.5601 0.5581						
A64	4	7 8 9	0.600 0.0849 0.0822	3.17 0.039 0.041	44.99 0.093 0.087	0.2305 0.2508	15.18 0.5493 0.5348	15.15 0.836 0.840	15.16 0.040 0.039	15.13 0.091 0.091	0.2419 0.2333	0.5457 0.5422						
A64	5	7 8 9	0.600 0.0917 0.0872	6.23 0.032 0.034	44.99 0.100 0.093	0.1776 0.1967	15.18 0.5479 0.5370	15.14 0.889 0.891	15.15 0.033 0.034	15.13 0.096 0.096	0.1886 0.1961	0.5451 0.5396						
A64	6	7 8 9	0.599 0.0877 0.0853	9.31 0.028 0.030	44.99 0.096 0.091	0.1618 0.1767	15.17 0.5473 0.5356	15.14 0.917 0.885	15.14 0.029 0.031	15.13 0.099 0.095	0.1631 0.1757	0.5437 0.5371						
A64	7	7 8 9	0.600 0.0907 0.0863	12.39 0.026 0.027	44.99 0.098 0.093	0.1461 0.1607	15.17 0.5446 0.5393	15.14 0.909 0.882	15.14 0.027 0.025	15.12 0.097 0.097	0.1537 0.1425	0.5381 0.5501						

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																
A64	8	0.600	0.0641	0.0041	44.99	0.2728	0.5117	15.14	15.14	15.15	15.13	0.2685	0.5653					
	8	0.0757	0.0041	0.0086	0.0086	0.2816	0.5626	0.0756	0.0038	0.0040	0.0084	0.2531	0.5652					
	9	0.0732	0.0041	0.0082	0.0082	0.2816	0.5626	0.0756	0.0038	0.0040	0.0085	0.2531	0.5652					
A65	1	0.599	-3.06	0.0008	44.99	0.2974	0.605	6.03	6.06	6.06	6.01	0.3365	0.6550					
	8	0.0139	0.0008	0.0020	0.0020	0.3707	0.6224	0.0142	0.0009	0.0009	0.0018	0.2330	0.6552					
	9	0.0117	0.0008	0.0014	0.0014	0.3707	0.6224	0.0147	0.0006	0.0006	0.0019	0.2330	0.6552					
A65	2	0.599	0.03	0.0018	44.99	0.2707	0.609	6.07	6.10	6.10	6.05	0.3020	0.6362					
	8	0.0343	0.0018	0.0043	0.0043	0.3640	0.6314	0.0324	0.0019	0.0019	0.0041	0.2605	0.6372					
	9	0.0286	0.0020	0.0038	0.0038	0.3640	0.6314	0.0329	0.0017	0.0017	0.0042	0.2605	0.6372					
A65	3	0.600	1.07	0.0023	44.99	0.2931	0.610	6.08	6.12	6.12	6.05	0.2970	0.6261					
	8	0.0403	0.0023	0.0052	0.0052	0.3139	0.6465	0.0381	0.0022	0.0022	0.0047	0.2629	0.6280					
	9	0.0372	0.0023	0.0043	0.0043	0.3139	0.6465	0.0388	0.0020	0.0020	0.0048	0.2629	0.6280					
A65	4	0.599	3.12	0.0031	44.99	0.2890	0.613	6.11	6.14	6.14	6.07	0.2967	0.6047					
	8	0.0538	0.0031	0.0065	0.0065	0.3197	0.6081	0.0493	0.0029	0.0029	0.0059	0.2723	0.6056					
	9	0.0496	0.0031	0.0058	0.0058	0.3197	0.6081	0.0491	0.0026	0.0026	0.0059	0.2723	0.6056					
A65	5	0.600	6.20	0.0037	44.99	0.2954	0.616	6.14	6.17	6.17	6.09	0.2765	0.5733					
	8	0.0673	0.0037	0.0068	0.0068	0.2822	0.5905	0.0660	0.0036	0.0036	0.0075	0.2730	0.5726					
	9	0.0656	0.0037	0.0068	0.0068	0.2822	0.5905	0.0627	0.0034	0.0034	0.0071	0.2730	0.5726					
A65	6	0.600	9.30	0.0041	44.99	0.2533	0.618	6.16	6.19	6.19	6.12	0.2648	0.5468					
	8	0.0809	0.0041	0.0087	0.0087	0.2896	0.5394	0.0796	0.0042	0.0042	0.0087	0.2497	0.5426					
	9	0.0718	0.0041	0.0081	0.0081	0.2896	0.5693	0.0753	0.0037	0.0037	0.0081	0.2497	0.5426					
A65	7	0.600	12.38	0.0039	44.99	0.2216	0.619	6.16	6.19	6.19	6.11	0.2265	0.5348					
	8	0.0883	0.0039	0.0086	0.0086	0.2230	0.5315	0.0833	0.0035	0.0035	0.0089	0.2138	0.5351					
	9	0.0817	0.0039	0.0086	0.0086	0.2230	0.5315	0.0833	0.0035	0.0035	0.0089	0.2138	0.5351					
A65	8	0.599	0.01	0.0020	44.99	0.2695	0.609	6.07	6.10	6.10	6.04	0.2970	0.6282					
	8	0.0342	0.0020	0.0038	0.0038	0.3508	0.6271	0.0321	0.0019	0.0019	0.0041	0.2599	0.6365					
	9	0.0292	0.0020	0.0038	0.0038	0.3508	0.6528	0.0324	0.0016	0.0016	0.0041	0.2599	0.6365					
A66	1	0.600	-3.12	0.0009	45.00	0.3992	-0.007	-0.09	0.00	0.00	-0.006	0.2606	0.6388					
	8	-0.0145	-0.0011	-0.0017	-0.0017	0.2390	0.6054	-0.0152	-0.0007	-0.0007	-0.0019	0.3562	0.6306					
	9	-0.0191	-0.0009	-0.0020	-0.0020	0.2390	0.5243	-0.0137	-0.0009	-0.0009	-0.0017	0.3562	0.6306					
A66	2	0.600	-0.03	0.0000	44.99	0.0148	-0.003	-0.06	0.05	0.05	-0.003	1.1317	-0.6577					
	8	0.0014	0.0000	0.0000	0.0000	-0.5824	1.2176	0.0005	0.0001	0.0001	-0.0000	-0.4117	-0.6899					
	9	-0.0016	0.0001	-0.0000	-0.0000	-0.5824	0.0571	0.0010	-0.0000	-0.0000	-0.0001	-0.4117	-0.6899					

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																						
866	3	7	0.601	0.98	44.99	0.2195	-0.02	-0.05	0.05	-0.02	0.0005	0.4077	0.5777	0.0070	0.0003	0.0010	0.3630	0.7244	-0.05	0.0003	0.0002	0.0008	0.1856	0.6895
A66	4	7	0.601	3.05	44.99	0.3144	-0.00	-0.03	0.09	-0.00	0.0009	0.3167	0.5902	0.0070	0.0011	0.0025	0.3661	0.6446	0.0153	0.0009	0.0007	0.0020	0.2414	0.6302
A66	5	7	0.601	6.16	44.99	0.3088	0.03	0.00	0.13	0.02	0.0042	0.2982	0.6046	0.0372	0.0023	0.0047	0.3048	0.6363	0.0350	0.0020	0.0018	0.0042	0.2720	0.6253
866	6	7	0.600	9.26	44.99	0.2870	0.07	0.04	0.17	0.06	0.0061	0.3006	0.5783	0.0566	0.0032	0.0066	0.3272	0.5849	0.0534	0.0032	0.0029	0.0062	0.3006	0.5783
A66	7	7	0.600	12.35	44.99	0.2692	0.10	0.07	0.19	0.09	0.0077	0.2682	0.5571	0.0718	0.0038	0.0078	0.3165	0.5840	0.0691	0.0039	0.0035	0.0074	0.2682	0.5571
A66	8	7	0.601	-0.04	44.99	0.2964	-0.04	-0.06	0.04	-0.03	-0.0001	0.6594	-0.5016	0.0009	0.0000	0.0003	-0.0002	1.7229	1.9017	0.0007	0.0001	-0.0001	-0.4445	-0.5016
867	6	7	1.250	-3.12	-0.00	-0.2894	-0.02	-0.11	0.05	-0.07	-0.0035	0.1026	0.6648	0.0020	0.0001	0.0002	-0.6496	0.9462	-0.0270	-0.0008	-0.0034	0.1026	0.6651	
867	7	7	1.250	-0.03	-0.00	-0.1377	-0.02	-0.06	0.05	-0.03	-0.0001	1.0670	-0.5733	0.0028	0.0000	0.0005	-1.0106	0.9801	0.0006	0.0001	0.0000	1.0670	-0.5733	
867	8	7	1.250	1.03	0.00	-0.0940	0.02	0.04	0.05	-0.01	0.0013	0.1974	0.6804	0.0028	0.0001	0.0002	10.0324	16.6465	-0.0080	0.0000	0.0000	0.1974	0.6804	
867	9	7	1.251	3.14	0.00	-0.1416	-0.02	-0.01	0.06	0.02	0.0038	0.1319	0.6636	0.0027	0.0002	0.0004	-90.6274	5.4965	0.0289	0.0007	0.0036	0.1319	0.6636	
867	10	7	1.252	6.27	0.00	-0.3604	-0.02	0.03	0.06	0.05	0.0073	0.1196	0.6622	0.0016	0.0001	0.0001	-0.7201	0.7201	0.0554	0.0013	0.0011	0.1196	0.6622	

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key													
869	1	1.249	-2.99	-0.00	-0.4242	-0.8287	15.09	0.06	15.07	0.1041	0.6808				
		0.0018	-0.0001	0.0003	0.4242	0.8287	0.0598	0.0012	0.0081	0.0923	0.6731				
		-0.0024	0.0003	-0.0002	-0.6563	0.4327	0.0603	0.0011	0.0081						
869	2	1.249	0.10	-0.00	-0.2623	-0.6771	15.13	0.06	15.11	0.0984	0.6504				
		0.0030	-0.0001	0.0004	0.2623	0.6771	0.0852	0.0016	0.0110	0.0908	0.6458				
		-0.0015	0.0002	-0.0001	-0.9630	0.5017	0.0846	0.0015	0.0109						
869	3	1.251	1.13	-0.00	-0.2007	-0.6571	15.14	0.06	15.12	0.0946	0.6389				
		0.0032	-0.0003	0.0004	0.2007	0.6571	0.0932	0.0017	0.0119	0.0877	0.6352				
		-0.0011	0.0003	-0.0001	-1.3104	0.5394	0.0924	0.0016	0.0117						
869	4	1.249	3.24	-0.00	-0.2060	-0.4459	15.16	0.06	15.14	0.0899	0.6260				
		0.0029	-0.0001	0.0002	0.2060	0.4459	0.1064	0.0019	0.0133	0.0844	0.6205				
		-0.0002	0.0003	-0.0000	-5.9357	0.8822	0.1058	0.0017	0.0131						
869	5	1.250	6.35	-0.00	-0.3445	-0.5772	15.19	0.07	15.15	0.1031	0.6000				
		0.0017	-0.0001	0.0002	0.3445	0.5772	0.1169	0.0024	0.0140	0.0988	0.5886				
		0.0001	0.0004	0.0002	20.3816	10.2042	0.1182	0.0023	0.0139						
869	6	1.249	9.47	-0.00	-0.2874	-0.6800	15.20	0.06	15.16	0.0883	0.5918				
		0.0016	-0.0000	0.0002	0.2874	0.6800	0.1290	0.0022	0.0152	0.0866	0.5822				
		-0.0010	0.0003	0.0000	-1.6882	-0.2457	0.1304	0.0022	0.0151						
869	7	1.249	12.68	0.00	-0.4012	-0.4807	15.21	0.06	15.17	0.0781	0.5859				
		0.0014	-0.0001	0.0001	0.4012	0.4807	0.1368	0.0021	0.0160	0.0780	0.5711				
		-0.0013	0.0003	-0.0000	-1.2531	0.3085	0.1360	0.0021	0.0155						
869	8	1.250	0.10	-0.00	-0.2153	-0.7906	15.13	0.06	15.10	0.0951	0.6452				
		0.0027	-0.0001	0.0004	0.2153	0.7906	0.0851	0.0016	0.0109	0.0911	0.6447				
		-0.0014	0.0003	-0.0001	-1.1569	0.4384	0.0841	0.0015	0.0108						
870	1	1.249	-2.95	44.99	0.1156	15.13	15.10	15.13	15.09	0.1074	0.6689				
		0.0683	0.0015	0.0093	0.1156	0.6879	0.0681	0.0014	0.0091	0.0909	0.6592				
		0.0645	0.0017	0.0085	0.1389	0.6581	0.0669	0.0012	0.0088						
870	2	1.249	0.16	44.99	0.1043	15.16	15.13	15.17	15.12	0.0965	0.6507				
		0.0858	0.0017	0.0113	0.1043	0.6596	0.0835	0.0016	0.0108	0.0960	0.6351				
		0.0824	0.0023	0.0103	0.1451	0.6253	0.0857	0.0016	0.0108						
870	3	1.250	1.20	44.99	0.0998	15.16	15.13	15.17	15.12	0.0940	0.6469				
		0.0914	0.0018	0.0118	0.0998	0.6495	0.0878	0.0016	0.0113	0.0976	0.6247				
		0.0867	0.0026	0.0107	0.1536	0.6179	0.0911	0.0017	0.0113						

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key															
870	4	7	1.249	3.26	44.99	0.0127	0.0996	15.18	15.15	15.19	15.14	0.0914	0.6396				
		9	0.1001	0.0019	0.0115	0.0115	0.1478	0.6347	0.0967	0.0017	0.0123	0.1195	0.5999				
		9	0.0955	0.0028	0.0115	0.0115	0.1478	0.6040	0.0957	0.0022	0.0114						
870	5	7	1.250	6.38	44.99	0.0139	0.0962	15.20	15.17	15.21	15.15	0.0868	0.6274				
		9	0.1113	0.0021	0.0124	0.0124	0.1394	0.6268	0.1094	0.0019	0.0137	0.1105	0.5815				
		9	0.1067	0.0029	0.0124	0.0124	0.1394	0.5843	0.1063	0.0023	0.0123						
870	6	7	1.249	9.51	44.99	0.0140	0.1012	15.21	15.18	15.22	15.17	0.0829	0.6175				
		9	0.1168	0.0023	0.0133	0.0133	0.1264	0.6002	0.1185	0.0019	0.0146	0.1006	0.5708				
		9	0.1151	0.0029	0.0133	0.0133	0.1264	0.5781	0.1162	0.0023	0.0132						
870	7	7	1.251	12.62	44.99	0.0146	0.0936	15.21	15.18	15.22	15.17	0.0902	0.5956				
		9	0.1237	0.0023	0.0133	0.0133	0.1203	0.5926	0.1212	0.0021	0.0144	0.0977	0.5621				
		9	0.1168	0.0028	0.0133	0.0133	0.1203	0.5722	0.1197	0.0023	0.0134						
870	8	7	1.249	0.17	44.99	0.0112	0.1012	15.15	15.13	15.16	15.12	0.0959	0.6502				
		9	0.0861	0.0017	0.0103	0.0103	0.1506	0.6551	0.0834	0.0016	0.0108	0.0957	0.6349				
		9	0.0816	0.0024	0.0103	0.0103	0.1506	0.6357	0.0856	0.0016	0.0108						
871	1	7	1.249	-3.07	44.99	0.0021	0.1242	6.05	6.02	6.07	6.02	0.1519	0.6709				
		9	0.0150	0.0003	0.0015	0.0015	0.3641	0.7092	0.0157	0.0004	0.0021	0.0656	0.6479				
		9	0.0113	0.0008	0.0015	0.0015	0.3641	0.6833	0.0139	0.0001	0.0018						
871	2	7	1.251	0.06	44.99	0.0050	0.0937	6.08	6.05	6.11	6.04	0.1045	0.6599				
		9	0.0371	0.0006	0.0047	0.0047	0.1721	0.6746	0.0350	0.0007	0.0046	0.0813	0.6595				
		9	0.0349	0.0012	0.0047	0.0047	0.1721	0.6845	0.0361	0.0005	0.0047						
871	3	7	1.250	1.10	44.99	0.0060	0.1100	6.10	6.06	6.12	6.05	0.0985	0.6681				
		9	0.0434	0.0009	0.0058	0.0058	0.1517	0.6924	0.0412	0.0008	0.0055	0.0861	0.6669				
		9	0.0441	0.0013	0.0058	0.0058	0.1517	0.6677	0.0437	0.0007	0.0058						
871	4	7	1.250	3.19	44.99	0.0076	0.1103	6.12	6.08	6.15	6.08	0.0947	0.6649				
		9	0.0564	0.0012	0.0079	0.0079	0.1538	0.6799	0.0535	0.0010	0.0075	0.0951	0.6570				
		9	0.0593	0.0018	0.0079	0.0079	0.1538	0.6701	0.0577	0.0010	0.0075						
871	5	7	1.250	6.32	44.99	0.0096	0.1041	6.15	6.11	6.19	6.11	0.1023	0.6508				
		9	0.0735	0.0015	0.0101	0.0101	0.1429	0.6571	0.0712	0.0014	0.0092	0.0969	0.6407				
		9	0.0786	0.0022	0.0101	0.0101	0.1429	0.6433	0.0774	0.0015	0.0099						
871	6	7	1.250	9.47	44.99	0.0112	0.1047	6.17	6.13	6.22	6.13	0.1003	0.6365				
		9	0.0871	0.0018	0.0117	0.0117	0.1341	0.6435	0.0850	0.0017	0.0108	0.0928	0.6258				
		9	0.0940	0.0025	0.0117	0.0117	0.1341	0.6243	0.0958	0.0017	0.0119						

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key																			
873	4	7	1.250	0.10	-0.004	-0.1182	-0.02	15.14	0.05	15.10	0.1030	0.6531	0.0022	-0.0000	0.0004	-0.9229	0.9783	0.0843	0.0017	0.0110	0.0919	0.6537
		8	0.0012	0.0002	-0.0001	-0.9229	0.7871	0.0841	0.0015	0.0110	0.0919	0.6537	-0.0012	0.0002	-0.0001	-0.9229	0.7871	0.0841	0.0015	0.0110	0.0919	0.6537
		9	-0.0012	0.0002	-0.0001	-0.9229	0.7871	0.0841	0.0015	0.0110	0.0919	0.6537	-0.0012	0.0002	-0.0001	-0.9229	0.7871	0.0841	0.0015	0.0110	0.0919	0.6537
873	5	7	1.250	1.14	-0.004	-0.0729	-0.02	15.15	0.06	15.11	0.0991	0.6435	0.0025	-0.0000	0.0004	-0.0729	0.8947	0.0922	0.0018	0.0118	0.0896	0.6435
		8	0.0025	-0.0000	0.0004	-0.0729	0.8947	0.0922	0.0018	0.0118	0.0896	0.6435	0.0025	-0.0000	0.0004	-0.0729	0.8947	0.0922	0.0018	0.0118	0.0896	0.6435
		9	-0.0010	0.0002	-0.0001	-1.1408	0.7906	0.0918	0.0016	0.0118	0.0896	0.6435	-0.0010	0.0002	-0.0001	-1.1408	0.7906	0.0918	0.0016	0.0118	0.0896	0.6435
873	6	7	1.250	3.25	-0.003	-0.0580	-0.02	15.18	0.06	15.13	0.0972	0.6263	0.0022	-0.0000	0.0003	-0.0580	0.7553	0.1049	0.0020	0.0131	0.0972	0.6263
		8	0.0022	-0.0000	0.0003	-0.0580	0.7553	0.1049	0.0020	0.0131	0.0972	0.6263	0.0022	-0.0000	0.0003	-0.0580	0.7553	0.1049	0.0020	0.0131	0.0972	0.6263
		9	-0.0006	0.0003	-0.0000	-2.2191	0.4086	0.1054	0.0018	0.0131	0.0864	0.6232	-0.0006	0.0003	-0.0000	-2.2191	0.4086	0.1054	0.0018	0.0131	0.0864	0.6232
873	7	7	1.251	6.36	0.002	-0.3585	-0.02	15.20	0.05	15.15	0.0994	0.6011	0.0013	0.0000	0.0002	-0.3585	0.8332	0.1161	0.0023	0.0139	0.0994	0.6011
		8	0.0013	-0.0000	0.0002	-0.3585	0.8332	0.1161	0.0023	0.0139	0.0994	0.6011	0.0013	0.0000	0.0002	-0.3585	0.8332	0.1161	0.0023	0.0139	0.0994	0.6011
		9	0.0011	0.0003	0.0001	1.3938	0.4782	0.1192	0.0021	0.0141	0.0906	0.5937	0.0011	0.0003	0.0001	1.3938	0.4782	0.1192	0.0021	0.0141	0.0906	0.5937
873	8	7	1.249	9.48	0.002	-0.2861	-0.02	15.21	0.06	15.16	0.0898	0.5923	0.0013	0.0000	0.0002	-0.2861	0.8932	0.1283	0.0023	0.0152	0.0898	0.5923
		8	0.0013	-0.0000	0.0002	-0.2861	0.8932	0.1283	0.0023	0.0152	0.0898	0.5923	0.0013	0.0000	0.0002	-0.2861	0.8932	0.1283	0.0023	0.0152	0.0898	0.5923
		9	-0.0006	0.0002	-0.0000	-1.8984	0.1558	0.1299	0.0023	0.0151	0.0897	0.5837	-0.0006	0.0002	-0.0000	-1.8984	0.1558	0.1299	0.0023	0.0151	0.0897	0.5837
873	9	7	1.250	12.65	0.001	-0.4062	-0.02	15.21	0.05	15.16	0.0781	0.5873	0.0014	0.0000	0.0001	-0.4062	0.5023	0.1346	0.0021	0.0156	0.0781	0.5873
		8	0.0014	-0.0001	0.0000	-0.4062	0.5023	0.1346	0.0021	0.0156	0.0781	0.5873	0.0014	0.0000	0.0001	-0.4062	0.5023	0.1346	0.0021	0.0156	0.0781	0.5873
		9	-0.0019	0.0003	0.0000	-0.8285	-0.0162	0.1355	0.0021	0.0154	0.0806	0.5713	-0.0019	0.0003	0.0000	-0.8285	-0.0162	0.1355	0.0021	0.0154	0.0806	0.5713
873	10	7	1.251	0.06	-0.003	-0.2483	-0.02	15.14	0.05	15.10	0.0994	0.6484	0.0024	-0.0001	0.0003	-0.2483	0.8105	0.0842	0.0016	0.0109	0.0994	0.6484
		8	0.0024	-0.0001	0.0003	-0.2483	0.8105	0.0842	0.0016	0.0109	0.0994	0.6484	0.0024	-0.0001	0.0003	-0.2483	0.8105	0.0842	0.0016	0.0109	0.0994	0.6484
		9	-0.0018	0.0003	-0.0001	-0.8162	0.3862	0.0838	0.0015	0.0108	0.0902	0.6497	-0.0018	0.0003	-0.0001	-0.8162	0.3862	0.0838	0.0015	0.0108	0.0902	0.6497
874	1	7	1.250	-3.07	-0.003	-0.4931	-0.02	6.00	0.05	6.00	0.2327	0.7346	0.0018	-0.0001	0.0003	-0.4931	0.8724	0.0058	0.0002	0.0008	0.2327	0.7346
		8	0.0018	-0.0001	0.0003	-0.4931	0.8724	0.0058	0.0002	0.0008	0.2327	0.7346	0.0018	-0.0001	0.0003	-0.4931	0.8724	0.0058	0.0002	0.0008	0.2327	0.7346
		9	-0.0012	0.0002	-0.0000	-0.8466	1.3160	0.0053	-0.0000	0.0008	-0.0050	0.7489	-0.0012	0.0002	-0.0000	-0.8466	1.3160	0.0053	-0.0000	0.0008	-0.0050	0.7489
874	2	7	1.250	0.03	0.004	-0.2644	-0.03	6.06	0.05	6.06	0.1127	0.6815	0.0026	-0.0001	0.0002	-0.2644	0.8525	0.0374	0.0008	0.0047	0.1127	0.6815
		8	0.0026	-0.0001	0.0002	-0.2644	0.8525	0.0374	0.0008	0.0047	0.1127	0.6815	0.0026	-0.0001	0.0002	-0.2644	0.8525	0.0374	0.0008	0.0047	0.1127	0.6815
		9	-0.0010	0.0002	-0.0000	-1.2665	0.9743	0.0350	0.0006	0.0047	0.0952	0.6698	-0.0010	0.0002	-0.0000	-1.2665	0.9743	0.0350	0.0006	0.0047	0.0952	0.6698
874	3	7	1.250	1.08	0.004	-0.2064	-0.02	6.08	0.06	6.07	0.1103	0.6747	0.0029	-0.0001	0.0004	-0.2064	0.7326	0.0480	0.0010	0.0064	0.1103	0.6747
		8	0.0029	-0.0001	0.0004	-0.2064	0.7326	0.0480	0.0010	0.0064	0.1103	0.6747	0.0029	-0.0001	0.0004	-0.2064	0.7326	0.0480	0.0010	0.0064	0.1103	0.6747
		9	-0.0006	0.0002	-0.0001	-2.0296	1.4147	0.0452	0.0008	0.0060	0.0952	0.6698	-0.0006	0.0002	-0.0001	-2.0296	1.4147	0.0452	0.0008	0.0060	0.0952	0.6698
874	4	7	1.248	3.18	0.003	-0.0962	-0.02	6.10	0.06	6.09	0.1076	0.6593	0.0023	-0.0000	0.0003	-0.0962	0.8059	0.0658	0.0014	0.0086	0.1076	0.6593
		8	0.0023	-0.0000	0.0003	-0.0962	0.8059	0.0658	0.0014	0.0086	0.1076	0.6593	0.0023	-0.0000	0.0003	-0.0962	0.8059	0.0658	0.0014	0.0086	0.1076	0.6593
		9	0.0002	0.0002	-0.0000	7.2310	-1.8094	0.0638	0.0012	0.0083	0.0977	0.6572	0.0002	0.0002	-0.0000	7.2310	-1.8094	0.0638	0.0012	0.0083	0.0977	0.6572

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Pd	See Key																																
874	5	7	1.247	6.33	0.00	-0.4092	-0.02	6.14	0.05	6.13	0.0999	0.6380	8	0.0013	-0.0001	0.0002	-0.4092	0.8105	0.887	0.0017	0.113	0.0999	0.6380	9	0.0005	0.0003	0.0001	3.4765	1.3592	0.0890	0.0016	0.0112	0.0999	0.6327
874	6	7	1.250	9.42	0.00	-0.2868	-0.02	6.17	0.05	6.15	0.0926	0.6169	8	0.0014	-0.0000	0.0002	-0.2868	0.7919	1.085	0.0020	0.133	0.0926	0.6169	9	-0.0006	0.0002	-0.0000	-2.0913	0.0993	0.1085	0.0018	0.0132	0.0926	0.6126
874	7	7	1.251	12.65	0.00	-0.3593	-0.02	6.19	0.05	6.16	0.0900	0.5985	8	0.0012	0.0000	0.0001	-0.3593	0.6578	1.206	0.0021	0.144	0.0900	0.5985	9	-0.0004	0.0002	-0.0001	-2.8668	1.4057	0.1210	0.0021	0.0142	0.0900	0.5889
874	8	7	1.249	0.02	0.00	-0.2374	-0.02	6.05	0.06	6.04	0.1074	0.6745	8	0.0024	-0.0001	0.0004	-0.2374	0.9186	0.371	0.0007	0.050	0.1074	0.6745	9	-0.0012	0.0003	-0.0001	-1.2578	0.7932	0.0343	0.0006	0.0046	0.1074	0.6731
875	1	7	1.250	-3.13	-0.00	-0.4557	-0.02	-0.11	0.05	-0.08	0.1276	0.6892	8	0.0020	-0.0001	0.0003	-0.4557	0.7985	-0.275	-0.0007	0.037	0.1276	0.6892	9	-0.0022	0.0002	-0.0002	-0.6423	0.5228	-0.0275	-0.0009	-0.0037	0.1276	0.6894
875	2	7	1.250	-0.02	0.00	-0.2941	-0.02	-0.05	0.05	-0.03	0.2956	0.4749	8	0.0028	-0.0001	0.0004	-0.2941	0.7975	0.014	0.0000	0.001	0.2956	0.4749	9	-0.0008	0.0002	-0.0002	-1.5201	1.3439	0.0002	-0.0002	-0.0000	0.2956	0.6029
875	3	7	1.249	1.04	0.00	-0.2087	-0.02	-0.04	0.05	-0.02	0.1444	0.6257	8	0.0030	-0.0001	0.0004	-0.2087	0.7476	0.111	0.0003	0.013	0.1444	0.6257	9	-0.0007	0.0002	-0.0001	-1.6233	0.9534	0.0083	-0.0000	0.0010	0.1444	0.6487
875	4	7	1.249	3.12	0.00	-0.0306	-0.02	-0.00	0.06	-0.00	0.1175	0.6475	8	0.0021	-0.0000	0.0004	-0.0306	0.9849	0.310	0.0007	0.040	0.1175	0.6475	9	-0.0005	0.0003	0.0000	-2.8314	-0.3160	0.0284	0.0004	0.0036	0.1175	0.6421
875	5	7	1.249	6.27	0.00	-0.4693	-0.02	0.03	0.06	0.03	0.1090	0.6516	8	0.0014	-0.0001	0.0001	-0.4693	0.6246	0.584	0.0012	0.076	0.1090	0.6516	9	0.0009	0.0003	0.0001	1.7852	0.8557	0.0580	0.0011	0.0075	0.1090	0.6506
875	6	7	1.250	9.41	0.00	-0.3655	-0.02	0.07	0.05	0.07	0.1021	0.6370	8	0.0015	-0.0001	0.0001	-0.3655	0.6221	0.831	0.0016	0.105	0.1021	0.6370	9	0.0002	0.0002	-0.0001	-5.2341	-2.7205	0.0825	0.0015	0.0104	0.1021	0.6355
875	7	7	1.249	12.62	0.00	-0.3932	-0.02	0.10	0.06	0.09	0.0913	0.6191	8	0.0012	-0.0000	0.0001	-0.3932	0.5340	0.1033	0.0018	0.128	0.0913	0.6191	9	-0.0007	0.0002	-0.0001	-1.9535	0.7640	0.1029	0.0017	0.0126	0.0913	0.6143

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key															
875	8 9	1.248 0.0025 -0.0010	-0.0001 0.0002	-0.0004 -0.0002	-0.0004 -0.0002	-0.2548 -1.2963	-0.02 0.9403 0.9805	-0.06 0.0014 -0.0001	0.06 0.0000 -0.0002	-0.03 0.0000 -0.0000	0.1023 9.5036	0.2158 3.0189					
876	1 8 9	0.900 0.0023 -0.0014	-0.0001 0.0000 -0.0001	-0.0003 0.0004 -0.0003	-0.0003 0.0004 -0.0003	-0.1795 -0.6687	-0.02 0.8798 1.3247	-0.11 -0.0269 -0.0262	0.05 -0.0011 -0.0013	-0.06 -0.0035 -0.0034	0.2178 0.2659	0.6613 0.6503					
876	2 8 9	0.899 0.0040 -0.0010	-0.0001 0.0002	-0.0003 0.0005	-0.0003 0.0005	-0.1265 -1.0122	-0.02 0.6975 1.4660	-0.05 0.0023 0.0000	0.05 0.0001 -0.0000	-0.02 -0.0001 -0.0000	0.3696 -0.1820	0.3191 -5.8410					
876	3 8 9	0.899 0.0046 -0.0005	1.00 -0.0001 0.0001	-0.0003 0.0005	-0.0003 0.0005	-0.1573 -1.6326	-0.02 0.6145 3.2520	-0.04 0.0110 0.0081	0.05 0.0006 0.0002	-0.01 0.0014 0.0011	0.2718 0.1526	0.6681 0.7078					
876	4 8 9	0.898 0.0039 -0.0007	3.10 -0.0001 0.0002	0.0004 -0.0002	0.0004 -0.0002	-0.1430 -1.3787	-0.02 0.6139 1.8292	-0.00 0.0329 0.0286	0.05 0.0013 0.0010	0.01 0.0042 0.0037	0.2108 0.1825	0.6450 0.6538					
876	5 8 9	0.898 0.0003 -0.0013	6.19 0.0000 0.0003	-0.0000 0.0000	-0.0000 0.0000	0.4916 -1.1762	-0.02 0.5229 0.0946	0.03 0.0599 0.0597	0.06 0.0025 0.0023	0.05 0.0072 0.0073	0.2138 0.1954	0.6014 0.6147					
876	6 8 9	0.898 0.0001 -0.0006	9.26 -0.0000 0.0001	-0.0001 0.0000	-0.0001 0.0000	-0.3925 -1.1574	-0.02 1.9522 1.4240	0.05 0.0805 0.0814	0.05 0.0030 0.0030	0.08 0.0092 0.0094	0.1909 0.1858	0.5764 0.5784					
876	7 8 9	0.897 0.0012 -0.0013	12.41 -0.0001 0.0001	-0.0002 0.0000	-0.0002 0.0000	-0.6746 -0.4250	-0.02 0.8572 1.0121	0.07 0.0975 0.0987	0.04 0.0030 0.0029	0.09 0.0109 0.0110	0.1544 0.1516	0.5535 0.5584					
876	8 8 9	0.900 0.0039 -0.0012	-0.0001 0.0001 0.0001	-0.0003 0.0005	-0.0003 0.0005	-0.1577 -0.8169	-0.02 0.6911 1.2838	-0.06 0.0021 -0.0000	0.05 0.0001 -0.0000	-0.02 0.0001 -0.0000	0.3994 8.6423	0.3063 7.3502					
877	1 8 9	0.898 0.0021 -0.0015	-3.05 -0.0000 0.0001	-0.0004 0.0004	-0.0004 0.0004	-0.1520 -0.4741	-0.02 1.0972 1.3761	6.00 0.0081 0.0074	0.04 0.0006 0.0003	6.01 0.0012 0.0012	0.3821 0.2565	0.6996 0.8158					
877	2 8 9	0.897 0.0038 -0.0009	0.00 -0.0001 0.0001	-0.0005 0.0005	-0.0005 0.0005	-0.1442 -0.7514	-0.02 0.7729 1.6589	6.06 0.0400 0.0379	0.05 0.0019 0.0015	6.05 0.0053 0.0051	0.2386 0.2083	0.6670 0.6775					

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key																					
877	3	7	0.899	1.05	-0.0001	-0.0005	-0.1544	-0.6887	6.07	0.05	6.06	0.2336	0.6352	0.0040	-0.0001	0.0005	0.6887	0.0489	0.0022	0.0062	0.2029	0.6547	
		8	-0.0006				0.9551	2.6678	0.0470	0.0019	0.0061												
		9																					
877	4	7	0.898	3.14	0.0000	-0.0003	0.1352	-0.9666	6.10	0.05	6.09	0.2194	0.6019	0.0019	0.0003	0.0028	0.9666	0.644	0.0028	0.0077	0.2022	0.6069	
		8	-0.0019				-0.8739	1.0681	0.0636	0.0025	0.0077												
		9																					
877	5	7	0.896	6.23	-0.0002	0.0000	-0.2743	-0.7294	6.12	0.05	6.11	0.1670	0.5788	0.0007	0.0000	0.0031	0.7294	0.848	0.0031	0.0097	0.1828	0.5723	
		8	-0.0007				-4.2240	1.4726	0.0851	0.0031	0.0097												
		9																					
877	6	7	0.897	9.30	-0.0001	-0.0001	-0.6850	-0.6873	6.13	0.05	6.13	0.1493	0.5693	0.0014	0.0001	0.0029	0.6873	0.981	0.0029	0.0111	0.1464	0.5601	
		8	-0.0014				-0.7426	0.4659	0.1003	0.0029	0.0111												
		9																					
877	7	7	0.895	12.43	-0.0001	-0.0003	-2.3981	-1.7498	6.13	0.05	6.13	0.1164	0.5720	0.0004	0.0001	0.0025	1.7498	1.075	0.0025	0.0123	0.1204	0.5613	
		8	-0.0004				-0.4463	1.8190	0.1116	0.0025	0.0123												
		9																					
877	8	7	0.897	0.00	0.0002	-0.0005	-0.1650	-0.7228	6.06	0.05	6.05	0.2349	0.6607	0.0040	0.0002	0.0018	0.7228	0.401	0.0018	0.0053	0.2059	0.6749	
		8	-0.0040				-0.7167	0.8136	0.0377	0.0015	0.0053												
		9																					
878	1	7	0.898	-2.97	0.0002	-0.0003	-0.4335	-0.6881	15.09	0.05	15.08	0.2293	0.6184	0.0029	0.0002	0.0027	0.6881	0.606	0.0027	0.0074	0.2068	0.6184	
		8	-0.0029				-0.4242	1.0214	0.0616	0.0025	0.0074												
		9																					
878	2	7	0.897	0.06	-0.0001	-0.0003	-0.1790	-0.6500	15.12	0.06	15.10	0.2052	0.5814	0.0028	0.0001	0.0032	0.6500	0.796	0.0032	0.0092	0.1919	0.5843	
		8	-0.0028				-0.8705	1.9944	0.0809	0.0031	0.0092												
		9																					
878	3	7	0.899	1.08	0.0000	-0.0002	-0.1903	-1.0858	15.12	0.05	15.11	0.1850	0.5756	0.0020	0.0004	0.0031	1.0858	0.859	0.0031	0.0100	0.1790	0.5762	
		8	-0.0020				-0.9509	2.1086	0.0872	0.0031	0.0100												
		9																					
878	4	7	0.901	3.15	0.0000	-0.0003	-0.0600	-0.9285	15.12	0.05	15.11	0.1628	0.5687	0.0020	0.0003	0.0030	0.9285	0.938	0.0030	0.0106	0.1555	0.5690	
		8	-0.0020				-1.1016	1.7235	0.0946	0.0029	0.0106												
		9																					
878	5	7	0.899	6.23	-0.0001	-0.0002	-0.3557	-1.7114	15.13	0.05	15.11	0.1247	0.5710	0.0008	0.0002	0.0025	1.7114	1.014	0.0025	0.0115	0.1226	0.5653	
		8	-0.0008				-1.0803	-1.4729	0.1041	0.0025	0.0115												
		9																					

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd																			
878	6	7	0.898	9.31	0.002	-0.8779	-0.02	15.13	0.05	15.11	0.1158	0.5698								
	6	9	0.0007	-0.0001	0.0002	0.8779	1.6070	0.1118	0.0025	0.0127	0.1154	0.5698								
	9		-0.0015	0.0001	-0.0000	-0.6338	0.1911	0.1139	0.0026	0.0127										
878	7	7	0.898	12.44	0.002	-0.6682	-0.02	15.14	0.04	15.11	0.0985	0.5726								
	6	9	0.0009	-0.0001	0.0002	0.6682	1.2606	0.1201	0.023	0.0137	0.0929	0.5681								
	9		-0.0016	0.0001	-0.0002	-0.4318	0.6342	0.1215	0.0022	0.0138										
878	8	7	0.900	0.05	-0.002	-0.0823	-0.02	15.11	0.05	15.11	0.2025	0.5764								
	6	9	0.0025	-0.0000	0.0004	0.0823	0.7936	0.0793	0.032	0.0091	0.2025	0.5764								
	9		-0.0010	0.0001	-0.0002	-0.7676	1.2671	0.0803	0.0030	0.0093	0.1919	0.5791								
879	3	7	0.598	-2.97	-0.003	-0.2017	-0.02	15.11	0.06	15.10	0.2919	0.6053								
	6	9	0.0020	-0.0000	0.0003	0.2017	0.7259	0.0535	0.031	0.0064	0.2919	0.6053								
	9		-0.0021	0.0001	-0.0001	-0.2763	0.3508	0.0547	0.0029	0.0067	0.2681	0.6122								
879	4	7	0.598	0.01	-0.004	-0.0018	-0.02	15.15	0.06	15.13	0.2665	0.5661								
	6	9	0.0028	-0.0000	0.0004	0.0018	0.7677	0.0755	0.040	0.0085	0.2665	0.5661								
	9		-0.0018	0.0000	-0.0001	-0.2579	0.4755	0.0747	0.0038	0.0084	0.2563	0.5678								
879	5	7	0.598	1.08	-0.004	-0.0983	-0.02	15.15	0.05	15.13	0.2558	0.5526								
	6	9	0.0036	-0.0000	0.0004	0.0983	0.5909	0.0799	0.040	0.0088	0.2558	0.5526								
	9		-0.0024	0.0001	-0.0001	-0.2716	0.2378	0.0792	0.0039	0.0087	0.2471	0.5516								
879	6	7	0.599	3.12	-0.004	-0.0310	-0.02	15.15	0.05	15.14	0.2130	0.5469								
	6	9	0.0030	-0.0000	0.0004	0.0310	0.6657	0.0852	0.036	0.0093	0.2130	0.5469								
	9		-0.0013	0.0001	-0.0001	-0.3814	0.5774	0.0865	0.0037	0.0093	0.2161	0.5396								
879	7	7	0.598	6.24	-0.002	-0.2523	-0.02	15.14	0.06	15.13	0.1668	0.5462								
	6	9	0.0014	-0.0000	0.0002	0.2523	0.9628	0.0875	0.029	0.0095	0.1668	0.5462								
	9		0.0001	0.0001	-0.0001	-3.2613	-3.0251	0.0901	0.0031	0.0096	0.1731	0.5334								
879	8	7	0.599	9.25	-0.002	-0.1321	-0.02	15.14	0.06	15.12	0.1419	0.5501								
	6	9	0.0008	-0.0000	0.0002	0.1321	1.7292	0.0891	0.025	0.0098	0.1419	0.5501								
	9		-0.0005	0.0000	-0.0001	-0.5584	1.1837	0.0910	0.0024	0.0098	0.1338	0.5419								
879	9	7	0.599	12.44	0.002	-0.3363	-0.02	15.14	0.05	15.12	0.1312	0.5540								
	6	9	0.0010	-0.0000	0.0002	0.3363	1.1593	0.0944	0.024	0.0104	0.1312	0.5540								
	9		-0.0025	0.0001	-0.0000	-0.2581	0.1613	0.0958	0.0023	0.0105	0.1229	0.5482								
879	10	7	0.599	0.07	-0.005	-0.3370	-0.02	15.16	0.05	15.13	0.2667	0.5652								
	6	9	0.0021	0.0001	0.0005	0.3370	1.2854	0.0756	0.040	0.0085	0.2667	0.5652								
	9		-0.0008	0.0000	-0.0002	-0.2177	1.5303	0.0750	0.0038	0.0084	0.2541	0.5638								

See Key

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run Pt.	Cd	See Key													
880	1	7	0.598	-3.04	-0.0001	-0.0002	-0.2992	-0.4984	-0.02	6.01	0.0070	0.0005	6.00	0.3825	0.6375
		8	0.0024	-0.0001	0.0002	-0.0002	-0.1557	0.3265	0.4984	0.0070	0.0067	0.0002	0.0008	0.2152	0.7642
		9	-0.0023	0.0000	-0.0001	-0.0001	-0.1557	0.3265	0.4984	0.0067	0.0067	0.0002	0.0010		
880	2	7	0.598	-0.00	-0.0004	-0.0004	0.1518	-0.9699	-0.02	6.07	0.0331	0.0020	6.04	0.3055	0.6431
		8	0.0025	0.0000	0.0004	0.0004	0.1518	0.9699	0.02	0.0331	0.0317	0.0016	0.0042	0.2599	0.6395
		9	-0.0008	0.0000	-0.0002	-0.0002	-0.1851	1.4112	1.4112	0.0317	0.0317	0.0016	0.0040		
880	3	7	0.598	1.04	-0.0004	-0.0004	0.0088	-0.6713	-0.02	6.08	0.0424	0.0025	6.06	0.3054	0.6274
		8	0.0034	0.0000	0.0004	0.0004	0.0088	0.6713	0.02	0.0424	0.0409	0.0021	0.0053	0.2680	0.6316
		9	-0.0010	0.0000	-0.0001	-0.0001	-0.3394	0.8597	0.8597	0.0409	0.0409	0.0021	0.0051		
880	4	7	0.599	3.15	-0.0004	-0.0004	0.0789	-0.7837	-0.02	6.12	0.0571	0.0033	6.09	0.2971	0.5844
		8	0.0027	0.0000	0.0004	0.0004	0.0789	0.7837	0.02	0.0571	0.0559	0.0033	0.0066	0.2775	0.5894
		9	-0.0017	0.0001	-0.0000	-0.0000	-0.3894	0.2178	0.2178	0.0559	0.0559	0.0031	0.0065		
880	5	7	0.598	6.21	0.0000	0.0002	-0.1993	-0.7112	-0.02	6.15	0.0762	0.0041	6.11	0.2698	0.5509
		8	0.0020	-0.0000	0.0000	0.0002	-0.1993	0.7112	0.02	0.0762	0.0760	0.0038	0.0084	0.2542	0.5478
		9	0.0003	0.0000	-0.0000	-0.0000	1.2067	-1.2892	-1.2892	0.0760	0.0760	0.0038	0.0083		
880	6	7	0.598	9.30	-0.0001	-0.0002	-0.2473	-0.1479	-0.02	6.14	0.0867	0.0034	6.12	0.2005	0.5428
		8	0.0012	-0.0000	0.0000	0.0002	-0.2473	0.1479	0.02	0.0867	0.0876	0.0035	0.0094	0.2043	0.5343
		9	-0.0008	0.0001	-0.0000	-0.0000	0.6173	0.1479	0.1479	0.0876	0.0876	0.0035	0.0093		
880	7	7	0.598	12.41	-0.0000	-0.0002	-0.3665	-0.8338	-0.02	6.13	0.0860	0.0028	6.10	0.1628	0.5396
		8	0.0013	-0.0000	0.0000	0.0002	-0.3665	0.8338	0.02	0.0860	0.0876	0.0027	0.0092	0.1571	0.5316
		9	-0.0010	0.0000	-0.0000	-0.0002	0.0274	1.3159	1.3159	0.0876	0.0876	0.0027	0.0093		
880	8	7	0.597	-0.00	0.0000	0.0005	0.1915	-0.0472	-0.02	6.06	0.0339	0.0020	6.04	0.3021	0.6330
		8	0.0024	0.0000	0.0000	0.0005	0.1915	0.0472	0.02	0.0339	0.0320	0.0016	0.0042	0.2635	0.6458
		9	-0.0023	0.0001	-0.0000	-0.0000	-0.2959	0.1804	0.1804	0.0320	0.0320	0.0016	0.0041		
881	1	7	0.597	-3.10	-0.0000	-0.0002	-0.0892	-0.9563	-0.02	6.04	0.0233	0.0012	6.07	0.2627	0.6239
		8	0.0015	-0.0000	0.0000	0.0002	-0.0892	0.9563	0.02	0.0233	0.0220	0.0014	0.0029	0.3333	0.6285
		9	-0.0008	0.0000	-0.0000	-0.0003	0.1500	1.8250	1.8250	0.0220	0.0220	0.0014	0.0027		
881	2	7	0.598	-0.01	-0.0004	-0.0004	0.0115	-0.7336	-0.02	6.05	0.0013	0.0002	6.03	0.7481	0.1992
		8	0.0030	0.0000	0.0004	0.0004	0.0115	0.7336	0.02	0.0013	0.0013	0.0002	0.0000	-1.0293	0.6189
		9	-0.0011	0.0000	-0.0001	-0.0001	-0.1908	0.8497	0.8497	0.0005	0.0005	0.0001	0.0000		
881	3	7	0.599	0.99	-0.0004	-0.0002	0.0805	-0.7998	-0.02	6.04	0.0092	0.0006	6.01	0.3675	0.6253
		8	0.0029	0.0000	0.0004	0.0004	0.0805	0.7998	0.02	0.0092	0.0092	0.0006	0.0011	0.1937	0.6552
		9	-0.0010	0.0000	-0.0002	-0.0002	-0.1945	1.0028	1.0028	0.0075	0.0075	0.0002	0.0009		

TABLE VI
CANARD POSITIONS, FORCE AND MOMENT COEFFICIENTS - AERODYNAMIC PHASE

Run	Pt.	Cd	See Key →														
881	4	7	0.598	3.11	-0.00	0.0541	-0.02	-0.00	0.06	-0.00	0.0016	0.0032	0.3141	0.6163			
		8	0.0031	0.0000	0.0004	0.0541	0.6894	0.0267	0.0016	0.0032	0.0012	0.0030	0.2589	0.6202			
		9	-0.0011	0.0000	-0.0001	-0.3701	0.5434	0.0243	0.0012	0.0030	0.0012	0.0030	0.2589	0.6202			
881	5	7	0.598	6.17	-0.00	-0.3462	-0.02	0.04	0.06	0.06	0.0060	0.3071	0.5917				
		8	0.0022	-0.0001	0.0002	-0.3462	0.4629	0.0511	0.0031	0.0060	0.0061	0.2740	0.5917				
		9	-0.0004	0.0001	-0.0000	-1.3685	0.3882	0.0515	0.0028	0.0061	0.0061	0.2740	0.5917				
881	6	7	0.597	9.28	0.00	-0.1802	-0.02	0.08	0.05	0.08	0.0080	0.2793	0.5506				
		8	0.0009	-0.0000	0.0002	-0.1802	1.4704	0.0728	0.0040	0.0080	0.0079	0.2646	0.5506				
		9	0.0006	-0.0000	-0.0002	-0.3437	-2.4355	0.0725	0.0038	0.0079	0.0079	0.2646	0.5506				
881	7	7	0.598	12.40	-0.00	-0.4556	-0.02	0.09	0.05	0.09	0.0091	0.2156	0.5355				
		8	0.0015	-0.0001	0.0002	-0.4556	0.6547	0.0857	0.0036	0.0091	0.0092	0.2170	0.5355				
		9	-0.0025	0.0000	-0.0001	-0.1894	0.1970	0.0869	0.0037	0.0092	0.0092	0.2170	0.5355				
881	8	7	0.598	-0.06	-0.00	0.3731	-0.02	-0.06	0.05	-0.03	0.0000	0.6900	0.1868				
		8	0.0019	0.0001	0.0005	0.3731	1.3563	0.0014	0.0002	0.0000	0.0000	0.6900	0.1868				
		9	-0.0010	-0.0000	-0.0002	0.0125	1.3575	0.0005	-0.0000	0.0000	0.0000	-0.6480	0.1868				