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**AN EXPERIMENTAL INVESTIGATION  
OF BOUNDARY LAYER TRANSITION  
ON A CONE AT ANGLE OF ATTACK**

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Prepared for  
Space and Missile System Organization  
Norton Air Force Base, California

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**AN EXPERIMENTAL INVESTIGATION  
OF BOUNDARY LAYER TRANSITION  
ON A CONE AT ANGLE OF ATTACK**

**BY**

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**SEPTEMBER 1969**

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## FOREWORD

This report presents the results of work performed by the General Electric Re-entry and Environmental Systems Division during the second half of CY 1969 under Contract F04701-69-C-0116 which is jointly supported by SAMSO and ARPA. The work was accomplished under Task 7.8.2 (Asymmetric Transition Effects on Vehicle Trim) of the STREET G Program and was monitored by Lt. R. Padfield (U.S. Air Force, SAMSO) and K. Kresa (ARPA).

This report has been reviewed and is approved.

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## ABSTRACT

As the first part of the Strategic Re-entry Technology (STREET G) task to determine the effect of asymmetric transition on static stability, an experimental investigation of the influence of angle of attack on boundary layer transition front location and shape has been conducted at AEDC Tunnel B at Mach 8. The model was a stainless steel 7.2 degree half-angle cone with three bluntness ratios ( $R_N/R_B = 0.01, 0.02$ ). For each of the bluntness ratios, heat transfer data were obtained for four values of the free stream Reynolds number and for four angles of attack, that is,  $\alpha = 0$  degree, 1 degree, 2 degree, and 4 degree. Comparisons of the measured heat transfer at  $\alpha = 0$  degree were made with theory and good agreement resulted.

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## NOMENCLATURE

SYMBOLS	DEFINITION
$A_B$	Model base area, in. <sup>2</sup>
$C_m$	Pitching moment coefficient based on diameter
$C_p$	Specific heat BTU/lb <sup>o</sup> F
$d_B$	Model base diameter (11.00 in.)
$h$	Enthalpy
$k$	Thermal conductivity
$L$	Model length, in.
$M$	Mach number
$P$	Pressure, psia
$Pr$	Prandtl Number, $C_p\mu/k$
$q$	Dynamic pressure, psia
$q$	Heat transfer flux, BTU/ft <sup>2</sup> sec.
$r_o$	Local model outer radius
$R$	Universal gas constant, 1545.43 ft. lb. <sub>f</sub> <sup>o</sup> R lb <sub>m</sub> mole
$Re$	Reynolds number
$R_N/R_B$	Model nose bluntness ratio
$S_T$	Stanton number, $\dot{q}/(h_o - h_w)(\rho u)_\infty$
$s$	Model wetted length
$T$	Temperature
$t$	Time, seconds
$u$	Velocity component parallel to flow direction, fps.
$X$	Model axial distance, in.

### Greek Symbols

$\alpha$	angle of attack, degrees
$\gamma$	ratio of specific heats
$\mu$	absolute viscosity
$\rho$	density, lb <sub>m</sub> /ft <sup>3</sup>
$\tau$	model wall thickness (inches)

### Subscripts

e	edge conditions
o	stagnation conditions
w	wall conditions
$\infty$	free stream conditions

### Superscripts

*	reference conditions
---	----------------------

## SECTION I

### INTRODUCTION

As part of the Strategic Re-entry Technology Program (STREET G), experimental studies are being conducted to investigate the effects that asymmetric transition can have on the static stability and resulting trim angle of attack of an ablating re-entry vehicle. The experimental investigation is composed of two phases: the first, to map the transition front location and shape on a no-blowing conical model at angle of attack; the second, to measure the forces and moments on a porous model configured to simulate ablation in the asymmetric turbulent region determined in the first phase. From this test series, the objective is to establish whether the trim angles of attack that are observed in flight, beginning at the point of transition onset at the base, can be attributed to the effects of asymmetric transition progression along the body. The investigation will not duplicate the many phenomena that occur in flight but does represent a reasonable facsimile of the flow processes.

This report presents the experimental data obtained on the first phase, i.e., transition mapping on an impervious conical model, of the test program. The tests were conducted at Mach 8 in Tunnel B of the van Karman Facility at AEDC. Thermocouples have been utilized to establish the shape and location of the transition front as a function of angle of attack, Reynolds number, and nose bluntness.

## SECTION II

### MODEL DESCRIPTION AND TEST PROCEDURE

The model used in the heat transfer test was a 7.2 degree half-angle cone, 43.54 inches long, with a removable nose section, Figure 1. Three nose bluntness ratios were utilized for the model (i.e.,  $R_N/R_B = 0, 0.01, 0.02$ ). One hundred thermocouples were installed on the inside of the model wall. The concentration of thermocouples was on one-half the model with several installed on the opposite half to check the yaw symmetry of the heat transfer. Table I details the location and wall thickness at each thermocouple.

Data were obtained at Mach 8 for four (4) Reynolds numbers with the model pitched at angles of attack of 0 degree, 1 degree, 2 degree, 4 degree. A summary of the test conditions is presented in Table II. The four Reynolds numbers were chosen to place the end of the transition region at prescribed axial locations on the model at zero angle of attack. The desired end of transition locations were to span the model, from near the aft end to a point as far forward as possible. The sharp cone ( $R_N/R_B = 0$ ) and  $R_N/R_B = 0.02$  configurations were tested at each of the four Reynolds numbers while the 1 percent bluntness model was tested at the two lowest Reynolds numbers only.

The model was mounted on the tunnel injection mechanism and at each test condition was inserted into the tunnel at the desired angle of attack. Data were recorded on magnetic tape at a rate of two hundred samples per second per thermocouple. The model was withdrawn from the tunnel and cooled in the dump tank after obtaining data for each test condition.

TABLE 1. LOCATION AND WALL THICKNESS AT 100 THERMOCOUPLES

"X" Dim.	T/C No. on Each Ray and Wall Thk. in Inches															
	0° Ray	Wall Thk.	45° Ray	Wall Thk.	90° Ray	Wall Thk.	135° Ray	Wall Thk.	180° Ray	Wall Thk.	225° Ray	Wall Thk.	270° Ray	Wall Thk.	315° Ray	Wall Thk.
6.560	1	.049														
8.130	2	.0495														
10.988	3	.059	21	.059	39	.059	57	.059	75	.058						
12.841	4	.063	22	.064	40	.064	58	.064	76	.064						
14.694	5	.063	23	.060	41	.063	59	.063	77	.059						
16.547	6	.060	24	.060	42	.063	60	.062	78	.060						
18.400	7	.058	25	.058	43	.059	61	.060	79	.059						
20.253	8	.055	26	.056	44	.062	62	.058	80	.057						
22.106	9	.054	27	.053	45	.055	63	.051	81	.050						
23.959	10	.056	28	.054	46	.057	64	.053	82	.053						
25.812	11	.055	29	.055	47	.056	65	.053	83	.053						
27.665	12	.054	30	.054	48	.055	66	.053	84	.053						
29.518	13	.050	31	.049	49	.050	67	.048	85	.048						
31.523	14	.048	32	.047	50	.047	68	.047	86	.047			95	.045		
33.306	15	.048	33	.055	51	.052	69	.056	87	.048						
35.089	16	.056	34	.056	52	.052	70	.058	88	.052			96	.053		
36.872	17	.051	35	.054	53	.050	71	.055	89	.051						
38.655	18	.050	36	.0525	54	.049	72	.054	90	.046	93	.051	97	.051	99	.052
40.438	19	.050	37	.051	55	.048	73	.049	91	.049						
42.221	20	.048	38	.048	56	.048	74	.048	92	.046	94	.048	98	.048	100	.048

TABLE II. SUMMARY OF TEST CONDITIONS

$$M_{\infty} = 8$$

$Re_{\infty} \times 10^{-6}/ft$	$P_0$ PSIA	$T_0$ (°R)	$R_N/R_B$	$\alpha$ (Degrees)
3.79	860	1332.0	0, 0.02	0, 1, 2, 4
3.02	670	1310.0	0, 0.02	0, 1, 2, 4
2.20	470	1300.0	0, 0.01, 0.02	0, 1, 2, 4
1.32	280	1260.0	0, 0.01, 0.02	0, 1, 2, 4

## SECTION III

### TEST FACILITY DESCRIPTION

AEDC's hypersonic wind tunnel B is equipped with an axisymmetric contoured nozzle which provides Mach 6 and 8 flow through the 50-inch diameter test section. The tunnel is closed circuit which operates over a range of pressures with air supplied by a main compressor system. Stagnation temperatures sufficient to avoid liquefaction in the test section are obtained through the use of a natural-gas-fired combustion heater. The tunnel is equipped with a model injection system which facilitates model changes or cooling without requiring the tunnel to shut down.

Reynolds number variations from 0.30 to  $3.8 \times 10^6$  per foot correspond to dynamic pressure variations from 0.30 to 4.1 psi for a Mach number of 8.0. The stagnation temperature in the tunnel is approximately 900 degrees F. The model may be rolled 180 degrees and pitched a total of 30 degrees on a straight sting arrangement.

A more detailed description of the facility can be found in Reference 1.

## SECTION IV

### DATA REDUCTION

Data were recorded from the model sensors during the entire tunnel entry time period. That is, the temperature versus time history of each thermocouple was recorded at the rate of 200 samples per second. An initial time,  $t_0$ , was established when the model reached the tunnel centerline. Injection of the model from the dump tank below the tunnel to the centerline takes approximately 2 seconds. The data printout is based on this time as being zero. Since the aerodynamic heat transfer to the model is proportional to the slope of the temperature-time history at a prescribed time, a least square curve was fit to the  $\pm 10$  points in the vicinity of this time. The slope was then obtained from the curve fit. The data were calculated at four time intervals, 0.00, 0.25, 0.50, and 1.00 seconds from time zero. The graphical and tabulated data presented in this report are the  $t = 0$  data. A complete listing of all the available data may be obtained from the authors. The equations and curve fits employed at AEDC in reducing the Tunnel B and C thermocouple data may be found in Appendix A.

Some typical variations of the Stanton number ( $\sim$  heat transfer) with time for several thermocouples at three axial stations may be found in Figures 2 through 5. The relative accuracy of the heat transfer as calculated by the one-dimensional heat conduction equation

$$\dot{q} = \rho C_p \tau \frac{dT}{dt}$$

which assumes that there is no conduction within the model and that there is no convection, is approximately  $\sim 6$  percent. Roughly  $\pm 1$  percent can be attributed to the uncertainty in the specific heat of the material. The error bars shown in Figures 2-5 were drawn consistent with the aforementioned accuracy. Since the accuracy of the one-dimensional heat conduction equation degrades as the model wall is heated, due to conduction errors, the most accurate data are obtained as soon as the model responds to the sensible aerodynamic heating. For the model wall thickness of 50 to 60 mils, the most accurate time interval is during the first second from time zero. It can be noted from Figures 2-5 that the time dependency of the Stanton number is generally within the accuracy of the heat transfer measurement and consequently, only the data at time zero were plotted. The stated accuracy of the data refers to "good" thermocouple installations. There were several thermocouples on the model that were damaged prior to and during the test. These gages will show up in the axial plots as the spurious points which obviously do not agree with the data trends. No attempt was made to screen out these points.

The Stanton number variation with axial length is plotted in Figures 6 through 57. The corresponding tabulated data are listed before each figure. The data are grouped by model bluntness ratio ( $R_N/R_B = 0, 0.01, \text{ and } 0.02$ ) and within each bluntness set are grouped with the Reynolds number in a decreasing order.

## SECTION V

### RESULTS

The heat transfer distribution on the cone model was obtained along several conical rays, as was noted earlier. The data were obtained at Mach 8 for four values of the free stream Reynolds number. For each Reynolds number, data were obtained at four angles of attack, namely 0 degree, 1 degree, 2 degree, and 4 degree. A limited investigation on the effects of nose bluntness on the transition front location and shape with angle of attack was also obtained.

#### DATA COMPARISONS WITH THEORY

Comparisons of the data obtained at zero angle of attack were made with the results of the Viscous Interaction Zero Angle of Attack Drag (VIZAAD) Program.<sup>(2)</sup> For laminar flow, the heat transfer is calculated from Lee's laminar flow equation<sup>(3)</sup> modified by reference enthalpy conditions to account for the variation of flow properties across the boundary layer:

$$\dot{q}_{\text{LAM}} = \frac{0.354 \rho^* \mu^* u_e r_o (h_T - h_w)}{(\text{Pr})^{2/3} \left[ \int_0^s \rho^* \mu^* u_e r_o^2 ds \right]^{1/2}}$$

For turbulent flow, Walker<sup>(4)</sup> obtained a solution for the combined momentum and energy integral equations. This solution provides the following relation for the local turbulent heat transfer in terms of boundary layer outer edge properties and reference enthalpy conditions:

$$\dot{q}_{\text{TURB}} = \frac{0.0296 \rho_e^{0.2} \mu_e^{0.05} (\mu^*)^{0.2} (\rho^*)^{0.8} u_e r_o^{0.25} (h_T - h_w)}{\left[ \int_0^s \rho_e \mu_e \mu_e^{0.25} r_o^{1.25} ds \right]^{0.2}}$$

In general, solutions were obtained wherein the state of the boundary layer was either all laminar or all turbulent. The data comparisons with theory in terms of the Stanton number,  $S_T = \dot{q}/(\rho u_\infty (h_o - h_w))$ , for the sharp cone model are presented in Figures 58 and 59; for  $R_N/R_B = 0.01$  in Figure 60; for  $R_N/R_B = 0.02$  in Figures 61 and 62. From these figures, it can be noted that there is excellent agreement between the data and the theory in the laminar region. The data exhibits an increase through the transitional flow region to a peak which is higher than the "all turbulent" theory. The heat transfer then decays at a rate somewhat faster than the "all turbulent" theory. At sufficiently large distances from the end of transition, the theory and data are in good agreement. In summary, it can be stated that for a non-ablating body, the heat transfer predicted by the VIZAAD program is in good agreement with data for laminar flows and for turbulent flows in the region far removed from the end of transition.

A summary of the deduced transition onset and end points for  $\alpha = 0$  degree is tabulated in Table III. Also listed in this table are the local wetted length Reynolds number  $R_{e_{ST}}$ , the local momentum thickness Reynolds number,  $Re_{\theta_T}$ , and the edge Mach number at the beginning and end of transition obtained from the laminar VIZAAD theory. The data are plotted in Figure 63. As  $R_{e_{\infty}}$  is reduced, the transitional flow region enlarges for both  $R_N/R_B = 0$  and 0.02. A summary of the deduced end of transition points on the cone for  $R_N = 0$  may be found in Table IV, for  $R_N/R_B = 0.01$  in Table V, and for  $R_N/R_B = 0.02$  in Table VI. Graphical representations of the data for  $\phi = 0$  degree (leeward ray),  $\phi = 90$  degree, and  $\phi = 180$  degree (windward ray) may be found in Figures 64, 65, and 66 for  $R_B = 0, 0.01, \text{ and } 0.02$ , respectively. The spatial distribution of the end of transition for each of the bluntness ratios investigated may be found in Figures 67-69.

For the sharp cone, the end of transition at angle of attack tends to move slightly rearward from the zero alpha location on the windward ray, and forward on the leeward ray. In particular, the forward movement on the leeward ray is more pronounced than is the aft movement on the wind side (Figure 67). This trend is similar to the sharp cone phenomena experienced by other experimenters<sup>(5-9)</sup>. It should be pointed out that this trend implies that transition occurs at a significantly lower local Reynolds number on the lee side than the windward rays. This forward movement on the lee side has classically been attributed to the destabilizing influences of cross flow.

As the cone is blunted to a value of  $R_N/R_B = 0.01$  ( $R_N = 0.055''$ ), the transition point moves forward on both the windward and leeward rays, from the zero alpha point (Figure 68). However, the forward movement is greater on the lee side than on the windward side. This trend with bluntness is identical to that noted by Stetson and Rushton<sup>(7)</sup> for bluntness ratios of  $R_N/R_B = 0.04$  and  $0.08$  ( $R_N = 1/8''$  and  $1/4''$ ). Furthermore, this lee side forward - wind side aft was noted in flight for blunt vehicles. On the lateral side ( $\phi = 90$  degrees), the transition location is slightly aft of the zero angle of attack value.

In addition to the data obtained at a one percent bluntness ratio, data were also obtained for  $R_N/R_B = 0.02$  (Figure 69). For this bluntness ratio, a reversal occurs. Namely, the transition point moves further forward on the windward ray than on the leeward ray. In addition, the end of transition was noted to be furthest aft on the  $\phi = 45$  degree ray. To the author's knowledge, the data reported by Maddalon and Henderson<sup>(10)</sup> are the only other source of data that indicate a similar phenomena. It should be pointed out that the data attributed to this source by many authors does not actually appear in this reference (i.e., Reference 10) but may be found in the paper by Bertram and Beckwith<sup>(11)</sup>. These data are for a sharp 2.87 degree half angle at a 2 degree angle of attack cone at  $M = 20$  in helium. The definition of transition was obtained with a pitot tube recording impact pressures near the model surface. It is not clear at this time why this apparent reversal occurs.

The prediction of the heat transfer in the transitional and the ensuing turbulent flow regions is out of scope for the current effort and should be the subject of future study. However, the one case ( $R_e = 3.79 \times 10^6$  per foot and  $R_N/R_B = 0.02$ ), comparisons were made using the VIZAAD theory for mixed laminar-turbulent solutions. That is, for a prescribed "transition" point, the momentum thickness from the laminar solution,  $\theta_L$ , is set equal to the turbulent momentum thickness,  $\theta_T$ . Then, using the turbulent flow relations, the remaining viscous properties  $\delta^*$ ,  $\delta$ ,  $C_f$ ,  $\dot{q}$ , etc. are computed. The implication here is that the turbulent solution has an "effective" origin consistent with  $\theta_T$ . Two such solutions were made for  $(X_T/L) = 0.5$  (transition onset) and for  $X_T/L = 0.72$  (end of transition). The results are shown in Figure 61. The magnitude of the "heat transfer overshoot" is accurately predicted by this simple approach. Furthermore, the two solutions tend to bracket the experimentally measured heat transfer values in the turbulent flow region.

#### TRANSITION REGION DEFINITION

As stated earlier, the objective of the first test phase was to establish, quantitatively, the shape of the transition front as deduced from the surface heat transfer measurements. This was performed for several values of the free stream Reynolds number, angle of attack, and the nose bluntness. Based on these results, the second phase of testing (i.e., the force phase) will employ a porous model with the blowing schedule tailored to conform to the transition front shape. That is, the surface mass transfer will be zero in the laminar-transitional region and will be non-zero in the turbulent region. The blowing rate will be varied during the test so as to encompass vehicles with refractory heat shields as well as vehicles with teflon heat shields. The forces and moments that are attributed to the asymmetric transition front will therefore be determined experimentally. Although in an actual flight case the blowing (i.e., ablation) schedule is non-zero everywhere, the aforementioned configuration was deemed a reasonable facsimile of flight.

The physical locations of the transition onset and end points were deduced from the data in the following manner:

The onset point was established to be the point where the heat transfer (i.e., the Stanton number) is a minimum (e.g., see Figures 58 - 62). The end of transition was deduced to be the point where the heat transfer is a maximum. There is some question as to exactly where the end of transition actually occurs because the peak heat transfer overshoots the "all turbulent" theory. For this investigation, however, the maximum heat transfer point was chosen consistently. Since the blowing schedule will be tailored to the end of transition points, emphasis was placed on this data.

As is evident in the data plots of Section IV, the exact location where transition begins or ends is not always clear. Consequently, from the first estimates deduced from the data, a smoothing process was employed wherein the transition points were cross plotted with the angle of attack,  $\alpha$ , meridian angle,  $\phi$ , free stream unit Reynolds number,  $R_{e\infty}/ft.$ , and nose bluntness  $R_N$ . The faired values resulting from this smoothing process were then rechecked against the heat transfer data. In all cases the resultant transition points actually reflect, to within the accuracy of the data, the trends deduced from the heat transfer measurements. It was felt that this process eliminates only the inherent data scatter that can be attributed to experimental uncertainties. Since the basic data are reported, the reader can draw independent conclusions if desired.

TABLE III. SUMMARY OF DEDUCED TRANSITION ONSET AND  
END POINTS FOR  $\alpha = 0^\circ$

$\alpha = 0^\circ$   
 $M_\infty = 8$   
 $\theta_c = 7.2^\circ$        $L = 3.628'$

GP.	Onset						End				
	$Re_\infty$ ( $\times 10^{-6}$ )	$R_N/R_B$	$T_W/T_G$	Lam. VIZAAD			Lam. VIZAAD				
				$X_{TR}/L$	$Re_{ST}$ ( $\times 10^{-6}$ )	$Re_{\theta T}$	$M_e$	$X_{TR}/L$	$Re_{ST}$ ( $\times 10^{-6}$ )	$Re_{\theta T}$	$M_e$
1	3.79	0	.405	.19	3.93	726	6.76	.38	7.86	1065	6.76
11	3.02	0	.40	.264	4.36	784	6.76	.447	7.38	1039	6.76
22	2.18	0	.40	.36	4.29	790	6.75	.595	7.11	1028	6.76
32	1.35	0	.425	.504	3.63	730	6.74	.885	6.40	978	6.74
47	2.15	.01	.42	.504	5.867	968	6.77	.758	8.97	1181	6.81
42	1.36	.01	.432	.63	4.73	864	6.79	--	--	--	--
6	3.79	.02	.390	.504	8.25	1231	6.44	.72	13.36	1524	6.64
17	3.05	.02	.40	.504	6.85	1116	6.49	.76	11.69	1415	6.68
								.80	12.45	1456	6.69
27	2.20	.02	.40	.588	6.32	1053	6.62	.88	10.24	1309	6.74
37	1.35	.02	.432	.724	5.116	925	6.71	--	--	--	--

TABLE IV. END OF TRANSITION (X/L)

$$R_N = 0$$

$$\phi = 43.54^\circ$$

$Re_\infty \times 10^{-6}/ft$	$\alpha$	$0^\circ$	$45^\circ$	$90^\circ$	$135^\circ$	$180^\circ$
3.79 ↓	$0^\circ$	.380	.380	.380	.380	.380
	1	.338	.358	.390	.41	.415
	2	.295	.338	.415	.420	.425
	4	.252	.335	.420	.431	.431
3.05 ↓	0	.447	.447	.447	.447	.447
	1	.380	.41	.457	.479	.480
	2	.315	.380	.485	.495	.500
	4	.260	.465	.510	.515	.520
2.20 ↓	0	.595	.595	.595	.595	.595
	1	.465	.550	.605	.630	.635
	2	.375	.508	.635	.650	.660
	4	.330	.507	.678	.685	.690
1.35 ↓	0	.885	.885	.885	.885	.885
	1	.650	.847	.900	.940	.960
	2	.540	.847	.960	---	---
	4	.470	.929	---	---	---

TABLE V. END OF TRANSITION (X/L)

$R_N/R_B = 0.01$        $L = 43.54''$

$Re_{\infty} \times 10^{-6}/ft$	$a$	$\phi$				
		$0^\circ$	$45$	$90$	$135$	$180$
2.20 ↓ 1.35 ↓	$0^\circ$	.758	.758	.758	.758	.758
	1	.624	.765	.78	.765	.715
	2	.55	.678	.765	.724	.678
	4	.46	.635	.724	.724	.678
	0	1.03				
	1	.80	.924			
	2	.635	.888			
	4	.58	.847			

TABLE VI. END OF TRANSITION (X/L)

$R_N/R_B = 0.02$        $L = 43.54''$

$Re_{\infty} \times 10^{-6}/ft$	$a$	$\phi$				
		$0^\circ$	$45$	$90$	$135$	$180$
3.79 ↓ 3.05 ↓ 2.20 ↓	$0^\circ$	.724	.724	.724	.724	.724
	1	.712	.78	.69	.635	.60
	2	.635	.76	.67	.59	.55
	4	.508	.593	.55	.49	.465
	0	.785	.785	.785	.785	.785
	1	.765	.847	.75	.712	.705
	2	.678	.806	.70	.65	.64
	4	.55	.635	.635	.56	.53
	0	.88	.88	.88	.88	.88
	1	.85	.929	.889	.82	.80
	2	.765	.889	.806	.74	.724
	4	.624	.765	.730	.62	.580

## SECTION VI

### CONCLUSIONS

An experimental investigation was conducted at Mach 8 in Tunnel B of the von Karman Facility at AFDC. The objective of these tests was to establish the asymmetric shape of the transition front on a cone at angle of attack. The model, instrumented with 100 thermocouples was a 7.2 degree half angle stainless steel cone with three nose configurations (i.e.,  $R_N/R_B = 0, 0.01, \text{ and } 0.02$ ). For each configuration, data were obtained at four Reynolds numbers and for four angles of attack.

The transition front locations determined from this heat transfer phase indicated that the beginning of the turbulent region moved forward on the leeward side and aft on the windward meridian from the  $\alpha = 0$  degree locations for the sharp cone model. On the side meridian, a slight rearward shift occurred. For  $R_N/R_B = 0.01$ , the turbulent front moved forward of the  $\alpha = 0$  degree location on both the wind and lee meridians with the leeward side moving most forward. The side meridian location moved aft of the  $\alpha = 0$  degree value. For  $R_N/R_B = 0.02$ , an apparent reversal occurred. The leeward transition front was aft of the  $\alpha = 0$  degree location while the windward front was forward of this point. The maximum aft transition location occurred at  $\phi = 45$  degrees. This reversal is not presently explained. However, other available ground test data at larger bluntness ratios typical of current  $R/Vs$  ( $R_N > 0.25$ ) indicate trends similar to the sharp cone data obtained on the current test.

Comparisons of the  $\alpha = 0$  degree heat transfer data were made with the GE-VIZAAD theory. In general, the data in the laminar region agrees well with the theory. In the turbulent regime, good agreement was noted between an all turbulent VIZAAD theory and the data for a region far downstream of the end of transition. A heat transfer overshoot (over the all turbulent theory) was noted in the vicinity of the end of transition. An instantaneous jump from a laminar to a turbulent theory, matching momentum thicknesses at the "transition" onset or end point, results in good agreement between prediction of the maximum heat transfer and the data.

## SECTION VII

### RECOMMENDATIONS

The objective of this phase of the STREET G task "Asymmetric Transition Effects on Vehicle Trim" is to establish, through experimental measurements, characteristic transition front shapes for a vehicle at small angle of attack. The results of these measurements will then be used to pattern the blowing schedule on a porous model (i.e., simulated ablation). Force and moment measurements are then to be obtained for several values of the blowing parameter, free stream Reynolds number, and angle of attack.

With the exception of the  $R_N/R_B = 0.02$  results, the data obtained in this investigation agree with trends noted in available ground and flight test experiments. Furthermore, it was concluded that the sharp cone data will provide transition front shapes that will tend to maximize the incremental moment coefficient due to a skewed front. The force and moment data that will be obtained in Phase 2 of the current task will be used to establish whether the variations in static stability obtained in the wind tunnel are in agreement with trends measured in flight. That is, does the traversal of a skewed transition front produce a  $\Delta C_m$  variation with axial station similar to that shown in Figure 70? Therefore, it is recommended that data be obtained for the Reynolds numbers and transition front shapes noted in Figure 71. The lower three Reynolds numbers were selected to establish the shape of the  $\Delta C_m$  curve when the front lies aft of the center of gravity. The high Reynolds number case establishes the effect forward of the center of gravity.

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**APPENDIX A. GENERAL HEAT TRANSFER PROGRAM  
TUNNELS A, B, AND C  
VON KARMAN FACILITY, AEDC**

**GENERAL**

The program computations and print-out outlined in this memo will be used to reduce Tunnels A, B, and C heat transfer data, effective this date. Future program changes will be distributed to all recipients of this memo.

**NOMENCLATURE**

A	Constant
b	Model skin thickness, ft
c	Model skin specific heat, Btu/lb-°R
D	Model characteristic length, ft
E	Thermocouple output, mvolts
$h_T$	Heat transfer coefficient, Btu/ft <sup>2</sup> -sec-°R
$h_H$	Heat transfer coefficient, $\frac{\text{Btu/ft}^2\text{-sec}}{\text{Btu/lb}}$
H	Air enthalpy, Btu/lb
k	Model skin conductivity coefficient, Btu/ft-sec-°R
M	Mach number
p	Pressure, psia
q	Dynamic pressure, psia
Q	Heat transfer rate per unit area, Btu/ft <sup>2</sup> -sec
r	Model nose radius, ft
Re	Free-stream unit Reynolds number, ft <sup>-1</sup>
Re <sub>D</sub>	Reynolds number based on D
St	Stanton number
t	Time, sec.
T	Temperature, °R or °F
V	Velocity, ft/sec
w	Model skin density, lb/ft <sup>3</sup>
$\alpha$	Angle of attack, deg
$\beta$	Sideslip angle, deg
$\phi$	Roll angle, deg
$\psi$	Yaw angle, deg
$\mu$	Air viscosity, lb-sec/ft <sup>2</sup>
$\rho$	Air density, slug/ft <sup>3</sup>

### Subscripts

FR	Computed from Fay-Riddell theory
m	Model angle
o	Tunnel stagnation conditions
p	Sting prebend angle
s	Sector angle
w	Model wall conditions
∞	Tunnel free-stream conditions

### Superscript

' Conditions downstream of a normal shock

### IMPUTS

#### 1. Program Inputs

- a. Listed for each project number  
Project number  
Project engineer  
Data reduction times (maximum of 5 times, listed relative to centerline time)
- b. Listed for each model configuration  
Model configuration number

$$\left. \begin{array}{l} A_0 \\ A_1 \\ A_2 \\ A_3 \end{array} \right\} \text{where } c_1 = A_0 + A_1 T_w + A_2 T_w^2 + A_3 T_w^3 \\ \text{(model material 1)}$$

$$\left. \begin{array}{l} A_0^* \\ A_1 \\ A_2 \\ A_3 \end{array} \right\} \text{where } c_2 = A_0 + A_1 T_w + A_2 T_w^2 + A_3 T_w^3 \\ \text{(model material 2)}$$

Note: \*Zero indicates entire model is one material

$$\left. \begin{array}{l} A_0 \\ A_1 \\ A_2 \end{array} \right\} \text{where } k_1 = A_0 + A_1 T_w + A_2 T_w^2$$

$$\left. \begin{array}{l} A_0 \\ A_1 \\ A_2 \end{array} \right\} \text{where } k_2 = A_0 + A_1 T_w + A_2 T_w^2$$

r, ft\*

$$\left. \begin{array}{l} A_0 \\ A_1 \end{array} \right\} \text{Where } w_1 = A_0 + A_1 T_w$$

$$\left. \begin{array}{l} A_0 \\ A_1 \end{array} \right\} \text{where } w_2 = A_0 + A_1 T_w$$

- c. Listed for each thermocouple  
 Thermocouple number  
 Model material (1 or 2)

b, ft  
 D, ft\*\*

Notes: \*Zero indicates do not compute or tabulate  $St_{FR}$  or  
 or tabulate  $St/St_{FR}$   
 \*\*Zero indicates do not compute  $Re_D$  or tabulate  $St\sqrt{Re_D}$

## 2. Tape Inputs

<u>Position</u>	<u>Channel</u>	<u>Input</u>
00	100	Gp. no.
01	101	Run no.
02	102	Mach no.
03	103	$\alpha_p$
04	104	Date
05	105	Config.
06	119	$Q_c$ indicate bit
07	107	Time accumulator input, high
08	108	Time accumulator input, low

<u>Position</u>	<u>Channel</u>	<u>Input</u>
09	115	$\alpha_s$ input, high
10	116	$\alpha_s$ input, low
11	117	$\phi_s$ input, high
12	118	$\phi_s$ input, low
13	114	$p_o$ range indicate
14	106	Thermocouple switch position indicate
15	109	Spare
16	110	Spare
17	111	Spare
18	112	Spare
19	113	Spare
20	00	$p_o$ input
21	01	Model thermocouple
.	.	.
.	.	.
.	.	.
118	98	Model thermocouple
119	99	$T_o$ input

#### DATA REDUCTION PROCEDURE

##### 1. Compute for Each Data Reduction Time

$$\alpha_s^* = (\alpha_{\text{reading}} - \alpha_{\text{zero}})(1/300), \text{ deg}$$

$$\phi_s = (\phi_{\text{reading}} - \phi_{\text{zero}})(1/64), \text{ deg}$$

$$\alpha_m = \tan^{-1} (\tan \alpha_s \cos \phi_s) + \alpha_p, \text{ deg}$$

$$\psi_m = -\beta_m = -\sin^{-1} (\sin \alpha_s \sin \phi_s), \text{ deg}$$

$$p_o = (p_{o\text{reading}} - p_{o\text{zero}}) (\text{scale factor}), \text{ psia}$$

$$T_o^{**} = 128.485 + 44.8278E + 0.0811857E^2 - 0.0170464E^3 \\ + 0.000632876E^4 - 0.00000710569E^5 + 460, \text{ }^\circ\text{R}$$

$$H_o = 4.875 + 0.2235 T_o + 0.0000135 T_o^2, \text{ Btu/lb}$$

$$\mu_o = 2.270 T_o^{1.5} (T_o + 198.6)^{-1} \times 10^{-8}, \text{ lb-sec/ft}^2$$

$$M_o = \text{manual input, scan position 02, xx, xx}$$

$$T_{\infty}/T_0^{***} = (1 + 0.2 M_{\infty}^2)^{-1}$$

$$p_{\infty}/p_0^{***} = (1 + 0.2 M_{\infty}^2)^{-3.5}$$

$$p_0'/p_0^{***} = \left( \frac{6 M_{\infty}^2}{M_{\infty}^2 + 5} \right)^{3.5} \left( \frac{6}{7 M_{\infty}^2 - 1} \right)^{2.5}$$

$$T_{\infty} = \left( \frac{T_{\infty}}{T_0} \right) (T_0), \text{ } ^{\circ}\text{R}$$

$$p_{\infty} = \left( \frac{p_{\infty}}{p_0} \right) (p_0), \text{ psia}$$

$$p_0' = \left( \frac{p_0'}{p_0} \right) (p_0), \text{ psia}$$

$$q_{\infty} = 0.7 M_{\infty}^2 p_{\infty}, \text{ psia}$$

$$V_{\infty} = 49 M_{\infty}^2 (T_{\infty})^{0.5}, \text{ ft/sec}$$

$$\rho_{\infty} = 0.0839 p_{\infty} (T_{\infty})^{-1}, \text{ slugs/ft}^3$$

$$\mu_{\infty} = 8.051 \times 10^{-10} T_{\infty}, \text{ lb-sec/ft}^2$$

$$\text{Re}_{\infty} = \rho_{\infty} V_{\infty} (\mu_{\infty})^{-1}, \text{ ft}^{-1}$$

$$\text{St}_{\text{FR}} = \frac{0.254}{\rho_{\infty} V_{\infty} r^{0.5}} \frac{\mu_0^{0.4}}{T_0^{0.15}} (p_0')^{0.5}$$

$$\text{Wall temperature} = A_0 + A_1 E + A_2 E^2 + A_3 E^3$$

where, for Chromel-Alumel thermocouples

$T_w, ^\circ\text{F}$	$A_0$	$A_1$	$A_2$	$A_3$
0 to 270	132.159	43.5164	-0.295017	0.08131300
270 to 1000	122.453	47.1061	-0.237774	0.00377667
1000 to 1500	145.669	44.4543	-0.144298	0.00284639

or, for Iron-Constantan thermocouples

0 to 238	132.065	33.8251	-0.2213290	0.027791600
238 to 786	134.831	32.6871	-0.0250407	-0.000894075
786 to 1200	120.596	32.8380	0.0638280	-0.002162250

Notes: \*For Tunnel A

$$\alpha_s = (\alpha_{\text{reading}} - \alpha_{\text{zero}}) (1/100), \text{ deg}$$

\*\*For Tunnel A

$$T_o = -64.4605 + 141.503E - 38.2537E^2 + 7.18724E^3 - 0.680267E^4 + 0.0256285E^5 + 460, \text{ } ^\circ\text{R}$$

\*\*\*For Tunnel C

$$T_{\bullet}/T_o = (1 + 0.2 M_{\bullet}^2)^{-1} [0.9378 - 3.900 \times 10^{-6} p_o + (6533 + 0.6547 p_o - 0.4137 T_o - 0.0001354 p_o T_o) (10^{-8}) T_o]$$

$$p_{\bullet}/p_o = (1 + 0.2 M_{\bullet}^2)^{-3.5} [1.0562 + 49.57 \times 10^{-6} p_o - (3523 + 1.8300 p_o + 1.3839 T_o - 0.0002196 p_o T_o) (10^{-8}) T_o]$$

$$p_o' / p_o = \left( \frac{6 M_{\bullet}^2}{M_{\bullet}^2 + 5} \right)^{3.5} \left( \frac{6}{7 M_{\bullet}^2 - 1} \right)^{2.5} [1.0419 + 38.31 \times 10^{-6} p_o - (1968 + 0.7925 p_o + 1.6905 T_o) (10^{-8}) T_o]$$

2. Computed for Each Thermocouple at Each Data Reduction Time

$T_w$ , °R, and  $\frac{dT_w}{dt}$ , °R/sec, are computed from the center of a least-squares parabola through 21 consecutive wall temperature points.

$$Q = wbc \frac{dT_w}{dt}, \text{ Btu/ft}^2\text{-sec}$$

$$h_T = \frac{Q}{T_o - T_w}, \text{ Btu/ft}^2\text{-sec-}^\circ\text{R}$$

$$H_w = 4.875 + 0.2235 T_w + 0.0000135 T_w^2, \text{ Btu/lb}$$

$$h_H = \frac{Q}{H_o - H_w}, \frac{\text{Btu/ft}^2\text{-sec}}{\text{Btu/lb}}$$

$$St = \frac{h_H}{\rho_{\infty} V_{\infty}} \left( \frac{1}{32.17} \right)$$

$$\frac{St}{St_{FR}}$$

$$Re_D = Re_{\infty} D$$

$$St \sqrt{Re_D}$$

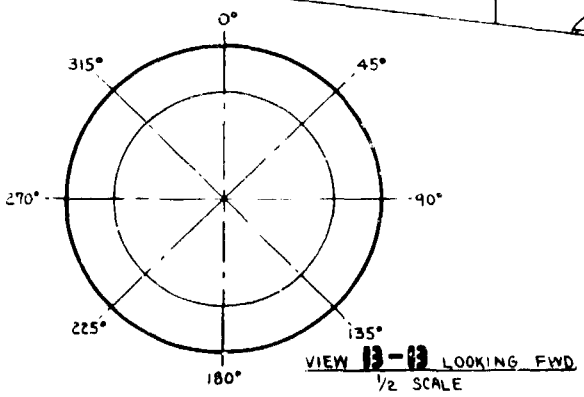
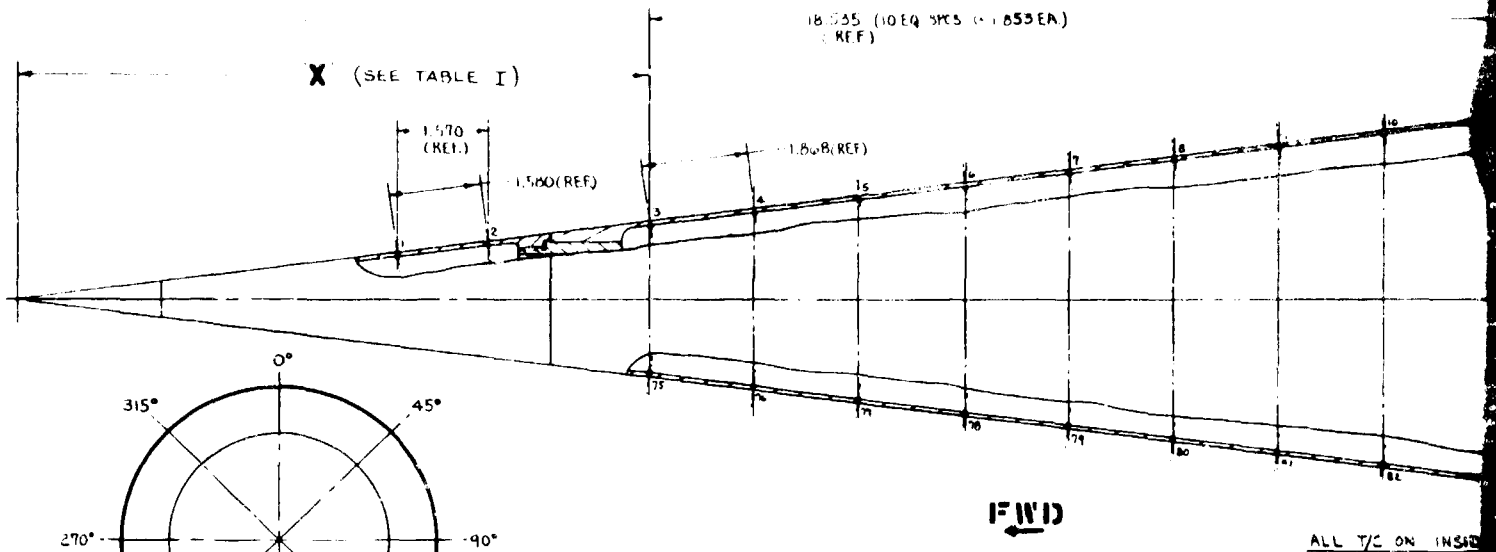
TABULATED DATA

1. Tabulated for Each Data Reduction Time

Gp.	$\alpha_g$	$\alpha_m$	$\phi_g$	$\psi_m$	Date	Config.	Time,	Mach	$p_o$	$T_o$
No.	deg	deg	deg	deg	_____	_____	sec.	No.	psia	$^{\circ}R$
$T_w$	$p_w$	$q_w$	$V_w$	$\rho_w$	$\mu_w$	$Re_w$	$St_{FR}$			
$^{\circ}R$	psia	psia	ft/sec	slug/ft <sup>3</sup>	lb-sec/ft <sup>2</sup>	ft <sup>-1</sup>	_____			

2. Tabulated for Each Thermocouple at Each Data Reduction Time

TC	$T_w$	$\frac{dT_w}{dt}$	$Q$	$H_T$	$h_H$	$H_w$
No.	$^{\circ}R$	$^{\circ}R/sec$	Btu/ft <sup>2</sup> -sec	Btu/ft <sup>2</sup> -sec- $^{\circ}R$	Btu/ft <sup>2</sup> -sec	Btu/lb
St	$\frac{St}{St_{FR}}$	$St \sqrt{Re_D}$				
_____	_____	_____				



1

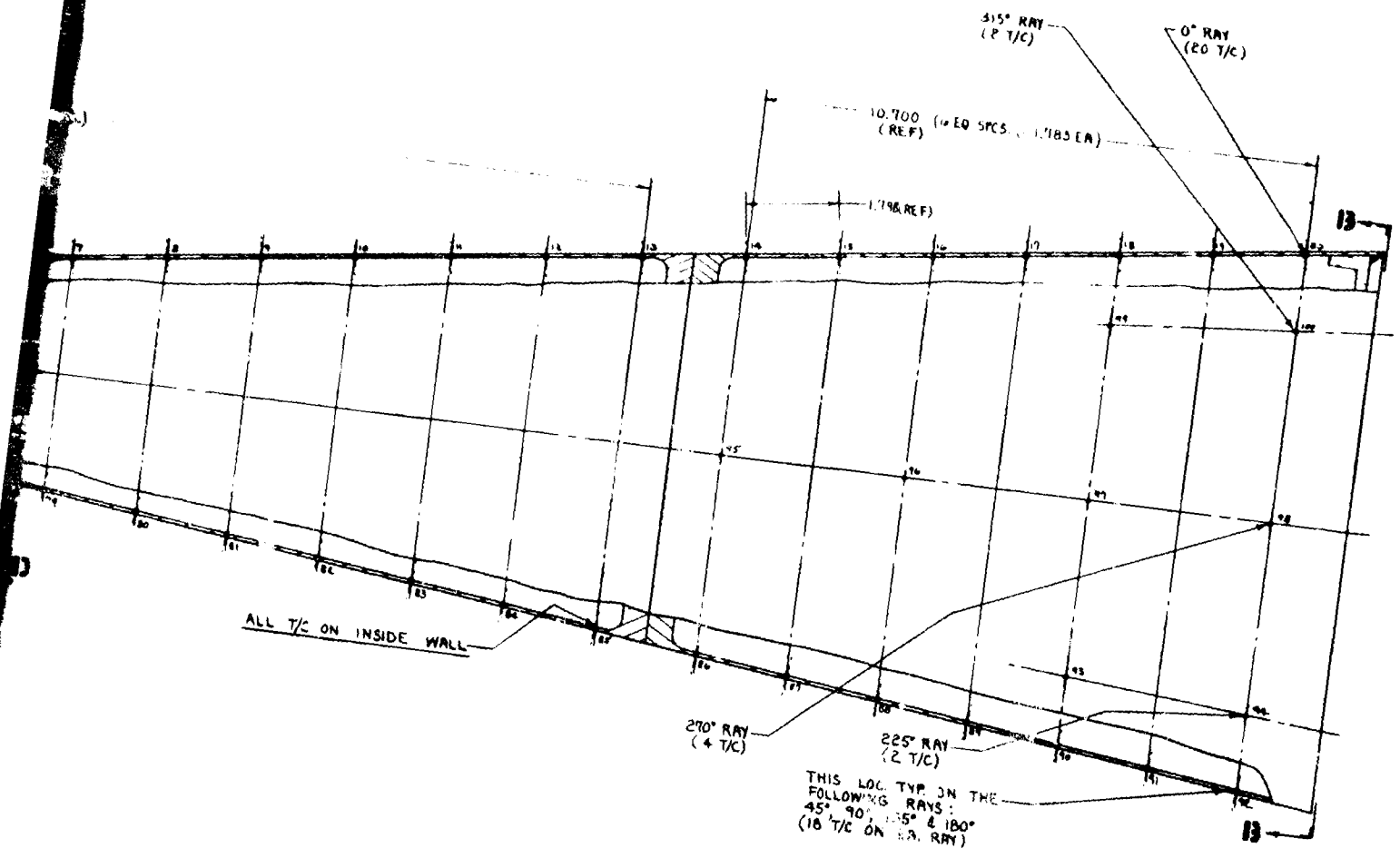


Figure 1. Model Used in Heat Transfer Test

*R*

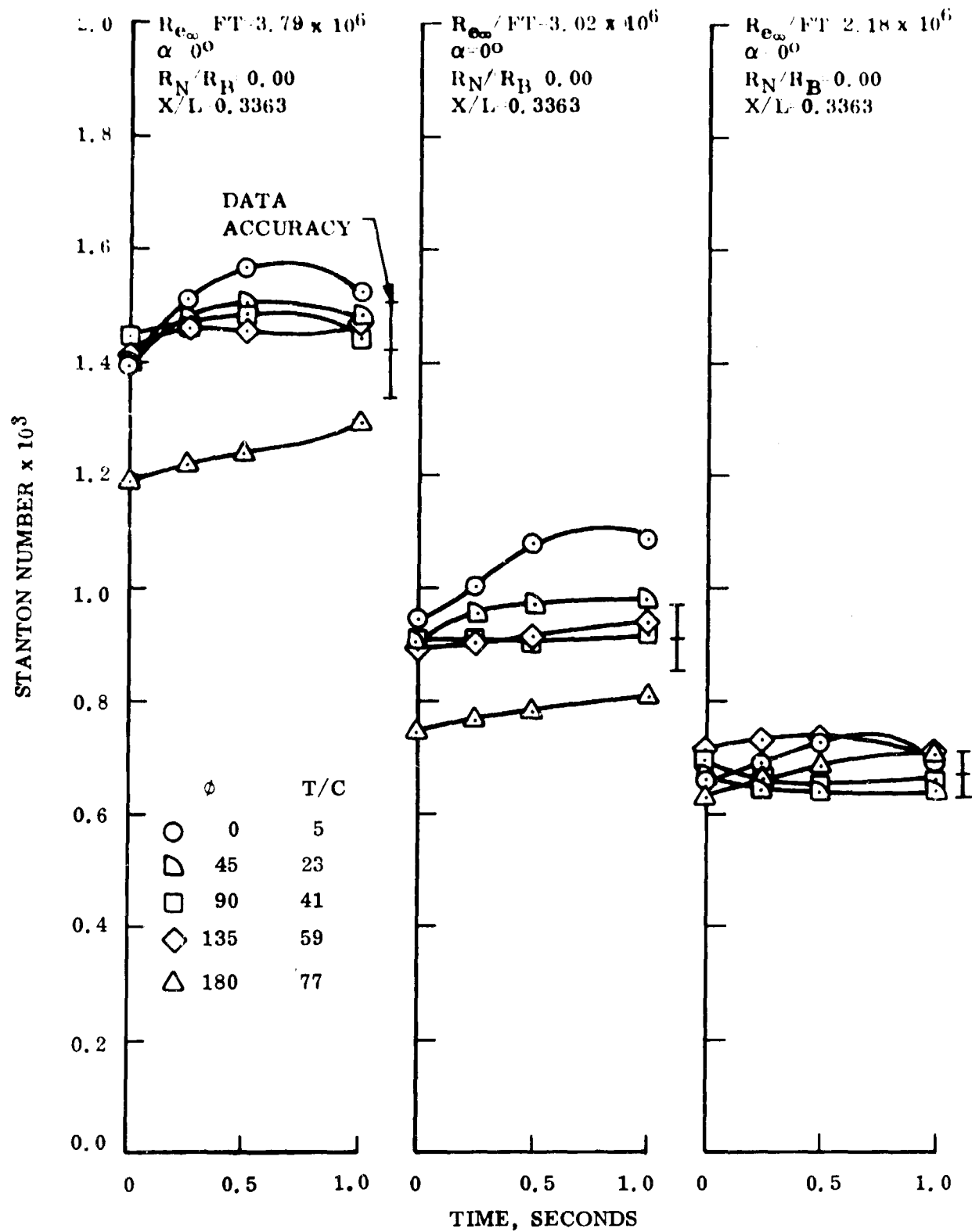


Figure 2. Stanton Number vs. Time for Selected  $X/L$  Locations,  $X/L = .3363$

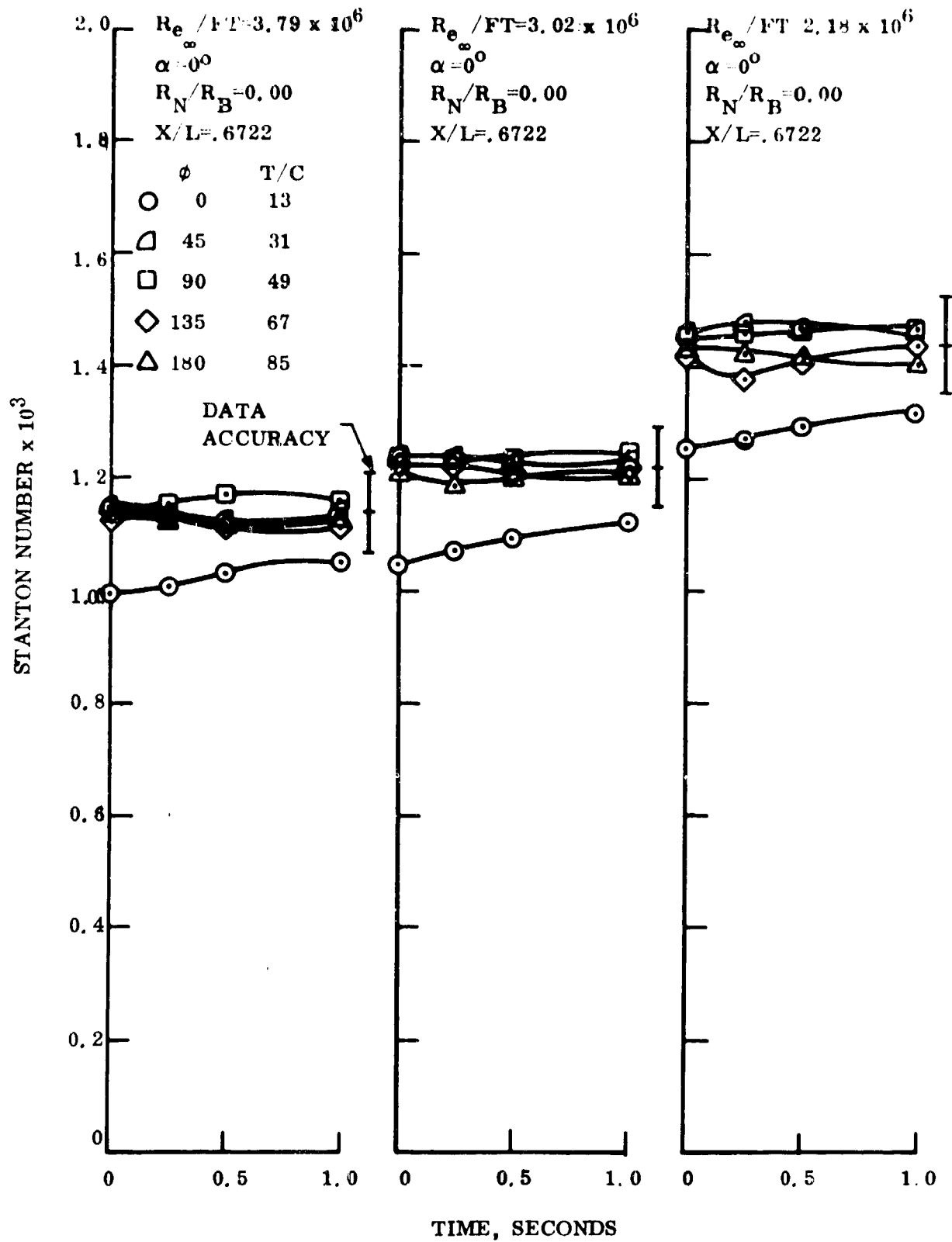


Figure 3. Stanton Number vs. Time for Selected X/L Locations, X/L = .6722

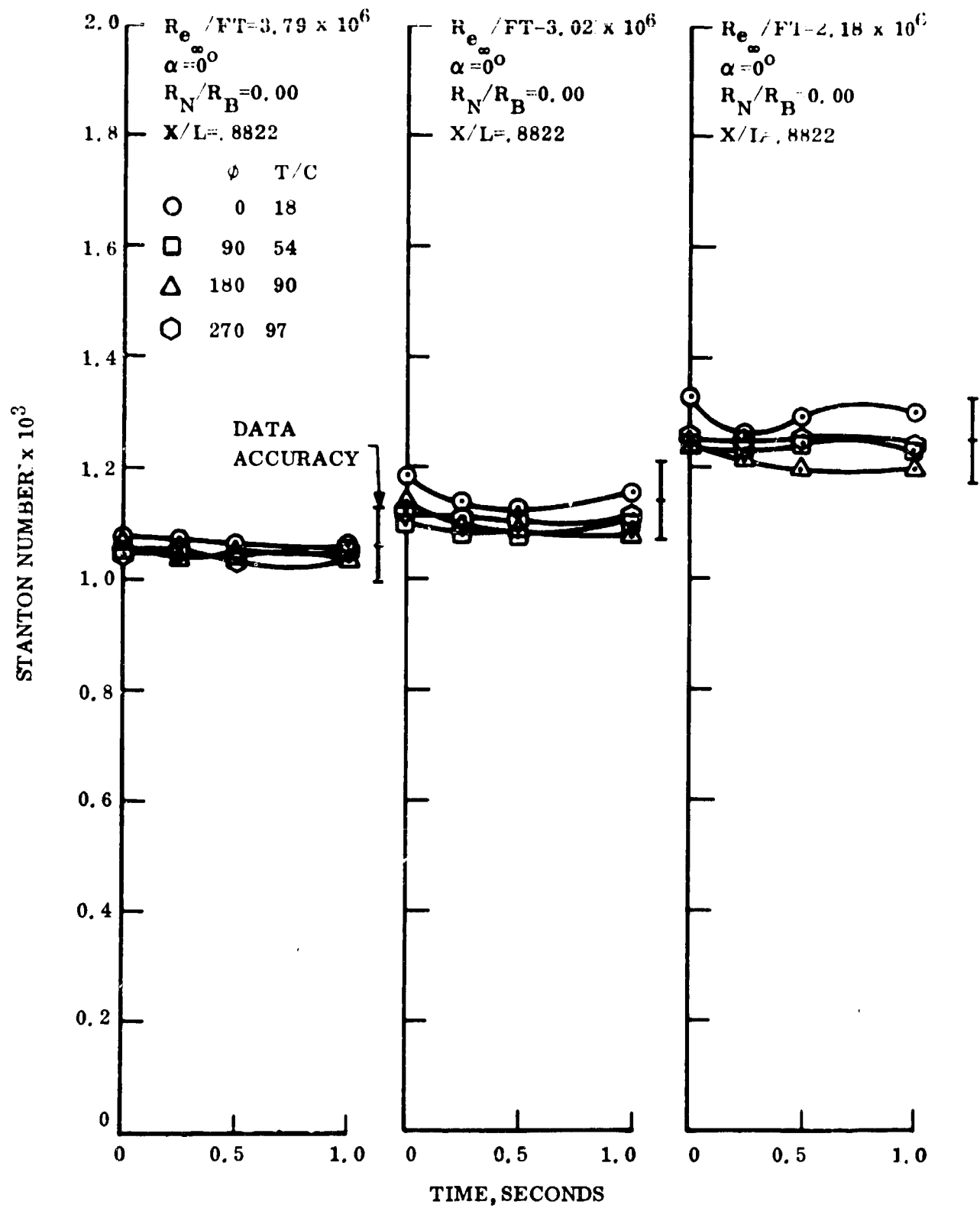


Figure 4. Stanton Number vs. Time for Selected X/L Locations, X/L = .8822

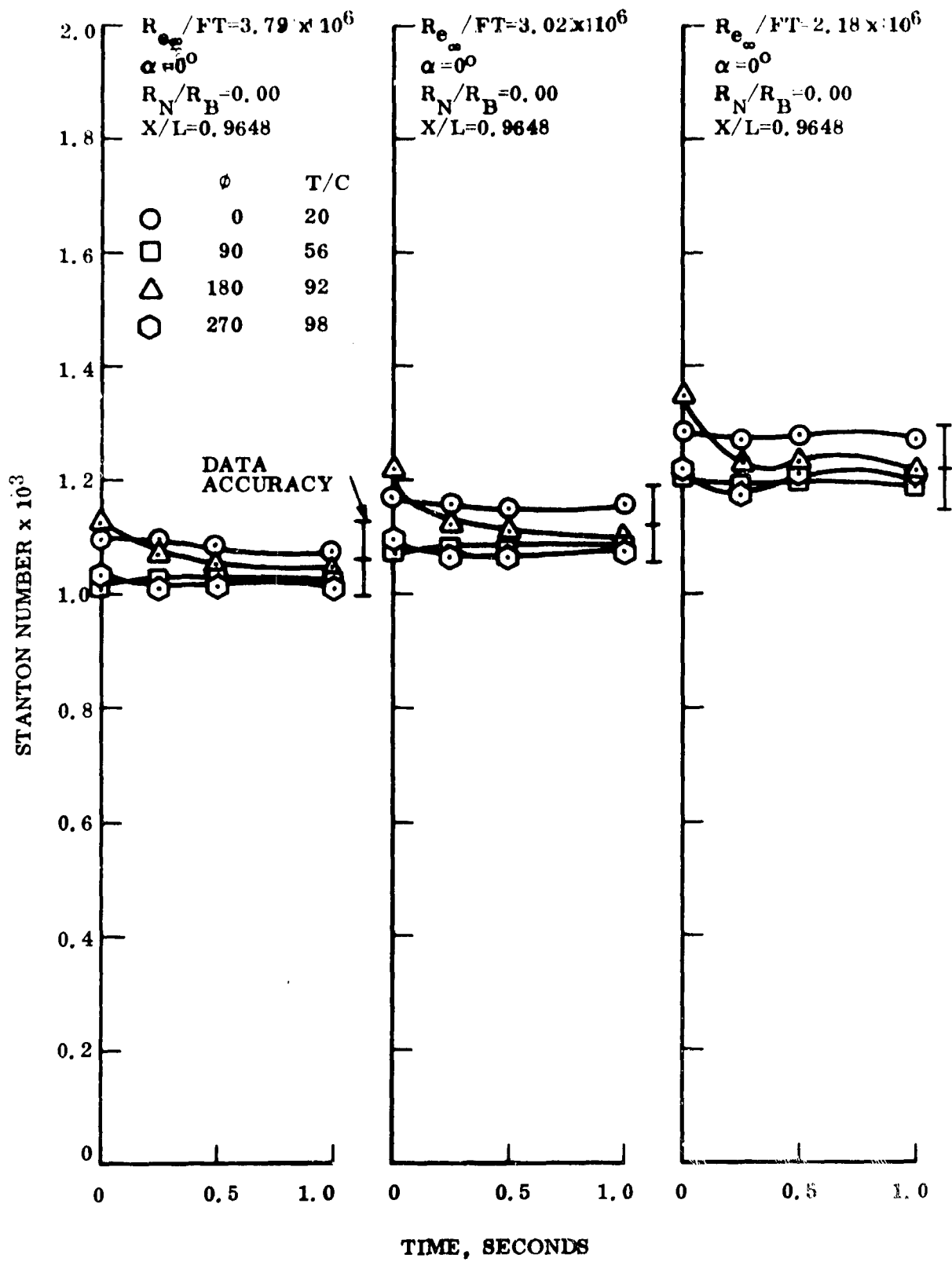


Figure 5. Stanton Number vs. Time for Selected X/L Locations. X/L = .9648

7/14/69

MEDICAL INC. ANNUL OF TENNESSEE  
VON KARMAN GAS DYNAMICS FACILITY  
DE INCH HYPERSONIC TUNNEL  
VT067-000

Table with columns: GROUP, CONFIG, ALPHA-N, BETA-H, ALPHA-S, NULL, ALP-P, DATE, MACH NO, PO, TO, TIME. Rows include data for various configurations and parameters like PC NO, X/L, PNE, TRMP, SLOPE, Q-STOP, HT, HM, MU, ST.

ASSEMBLY TRIM LOG  
HEB TRANSFER PHASE  
SYM PHT XAL SYP PHT XAL  
GROUP



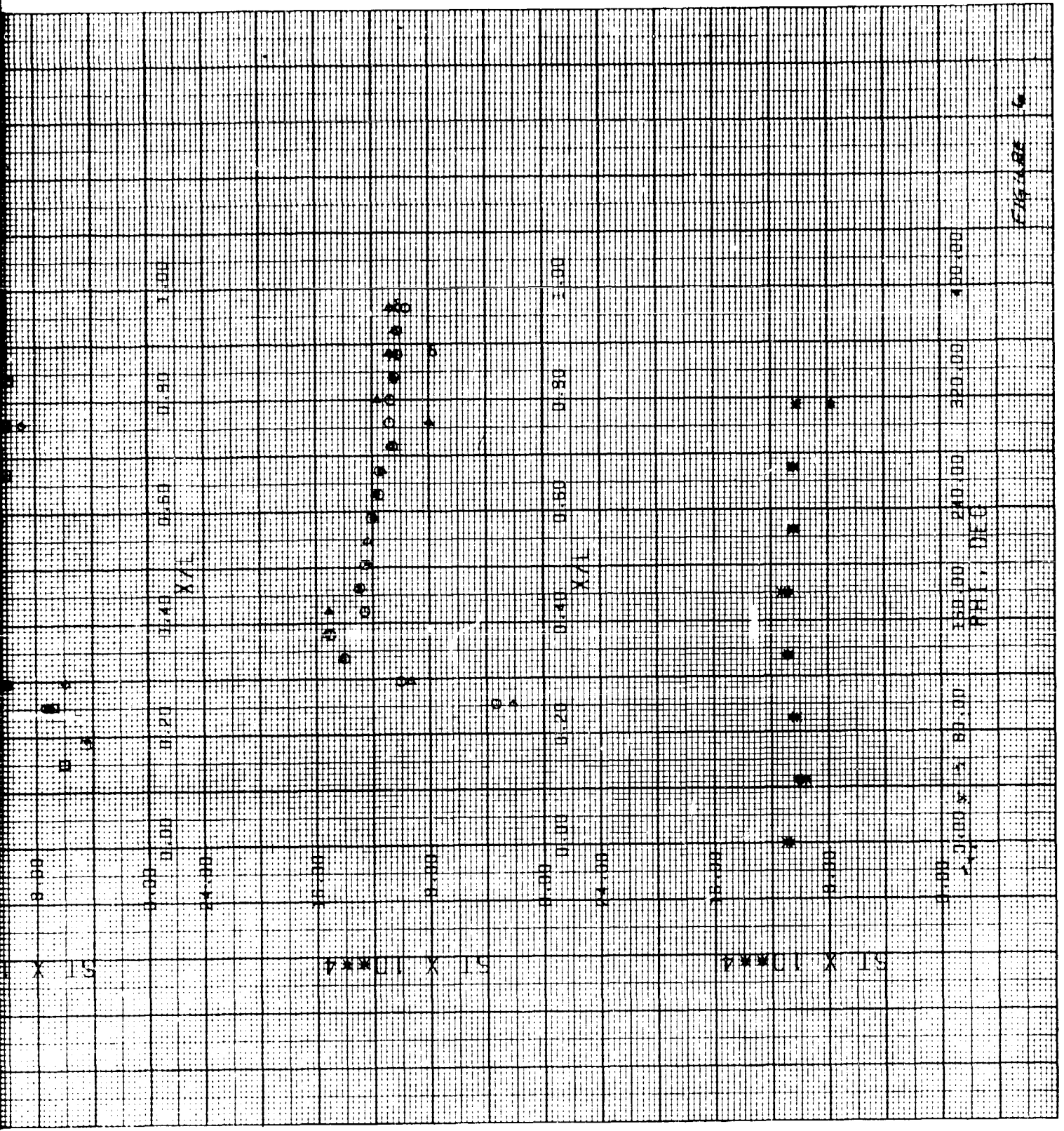


Figure 6

7/16/69

MECICARO INC. ANNUL OFS TENNESSEE  
VON KARMAN GAS DYNAMICS FACILITY  
50 INCH HYPERSONIC TUNNEL H

Table with columns: GROUP, CORR I, ALPHA-H, META-H, ALPHA-H, MULL, ALP-P, DATE, MACH NO, PO, TO, TIME. Rows include numerical data points for various parameters.

OF ASYMMETRIC TRIM TEST  
HEAD TRANSFER PHASE  
GROUP 2

GE ASYMMETRIC		RIM TEST	
HEAT	TRANSFER	PHASE	PHI
GROUP 2	ISYM	PHI	X/1
MACH NO. 8	0	0	541
RE. NO. 13.78 X 10**6	45	45	X
CONFID 1	90	90	X
ALPHA 1	135	135	0
BUMF 10 SEC	180	180	X
			0.88
			0.95
32-88			
34-88			
16-88			
8-88			
0.00	0.20	0.40	0.60
0.80	1.00	1.20	1.40
1.60	1.80	2.00	2.20
2.40	2.60	2.80	3.00
3.20	3.40	3.60	3.80
4.00	4.20	4.40	4.60
4.80	5.00	5.20	5.40
5.60	5.80	6.00	6.20
6.40	6.60	6.80	7.00
7.20	7.40	7.60	7.80
8.00	8.20	8.40	8.60
8.80	9.00	9.20	9.40
9.60	9.80	10.00	10.20
10.40	10.60	10.80	11.00
11.20	11.40	11.60	11.80
12.00	12.20	12.40	12.60
12.80	13.00	13.20	13.40
13.60	13.80	14.00	14.20
14.40	14.60	14.80	15.00
15.20	15.40	15.60	15.80
16.00	16.20	16.40	16.60
16.80	17.00	17.20	17.40
17.60	17.80	18.00	18.20
18.40	18.60	18.80	19.00
19.20	19.40	19.60	19.80
20.00	20.20	20.40	20.60
20.80	21.00	21.20	21.40
21.60	21.80	22.00	22.20
22.40	22.60	22.80	23.00
23.20	23.40	23.60	23.80
24.00	24.20	24.40	24.60
24.80	25.00	25.20	25.40
25.60	25.80	26.00	26.20
26.40	26.60	26.80	27.00
27.20	27.40	27.60	27.80
28.00	28.20	28.40	28.60
28.80	29.00	29.20	29.40
29.60	29.80	30.00	30.20
30.40	30.60	30.80	31.00
31.20	31.40	31.60	31.80
32.00	32.20	32.40	32.60
32.80	33.00	33.20	33.40
33.60	33.80	34.00	34.20
34.40	34.60	34.80	35.00
35.20	35.40	35.60	35.80
36.00	36.20	36.40	36.60
36.80	37.00	37.20	37.40
37.60	37.80	38.00	38.20
38.40	38.60	38.80	39.00
39.20	39.40	39.60	39.80
40.00	40.20	40.40	40.60
40.80	41.00	41.20	41.40
41.60	41.80	42.00	42.20
42.40	42.60	42.80	43.00
43.20	43.40	43.60	43.80
44.00	44.20	44.40	44.60
44.80	45.00	45.20	45.40
45.60	45.80	46.00	46.20
46.40	46.60	46.80	47.00
47.20	47.40	47.60	47.80
48.00	48.20	48.40	48.60
48.80	49.00	49.20	49.40
49.60	49.80	50.00	50.20
50.40	50.60	50.80	51.00
51.20	51.40	51.60	51.80
52.00	52.20	52.40	52.60
52.80	53.00	53.20	53.40
53.60	53.80	54.00	54.20
54.40	54.60	54.80	55.00
55.20	55.40	55.60	55.80
56.00	56.20	56.40	56.60
56.80	57.00	57.20	57.40
57.60	57.80	58.00	58.20
58.40	58.60	58.80	59.00
59.20	59.40	59.60	59.80
60.00	60.20	60.40	60.60
60.80	61.00	61.20	61.40
61.60	61.80	62.00	62.20
62.40	62.60	62.80	63.00
63.20	63.40	63.60	63.80
64.00	64.20	64.40	64.60
64.80	65.00	65.20	65.40
65.60	65.80	66.00	66.20
66.40	66.60	66.80	67.00
67.20	67.40	67.60	67.80
68.00	68.20	68.40	68.60
68.80	69.00	69.20	69.40
69.60	69.80	70.00	70.20
70.40	70.60	70.80	71.00
71.20	71.40	71.60	71.80
72.00	72.20	72.40	72.60
72.80	73.00	73.20	73.40
73.60	73.80	74.00	74.20
74.40	74.60	74.80	75.00
75.20	75.40	75.60	75.80
76.00	76.20	76.40	76.60
76.80	77.00	77.20	77.40
77.60	77.80	78.00	78.20
78.40	78.60	78.80	79.00
79.20	79.40	79.60	79.80
80.00	80.20	80.40	80.60
80.80	81.00	81.20	81.40
81.60	81.80	82.00	82.20
82.40	82.60	82.80	83.00
83.20	83.40	83.60	83.80
84.00	84.20	84.40	84.60
84.80	85.00	85.20	85.40
85.60	85.80	86.00	86.20
86.40	86.60	86.80	87.00
87.20	87.40	87.60	87.80
88.00	88.20	88.40	88.60
88.80	89.00	89.20	89.40
89.60	89.80	90.00	90.20
90.40	90.60	90.80	91.00
91.20	91.40	91.60	91.80
92.00	92.20	92.40	92.60
92.80	93.00	93.20	93.40
93.60	93.80	94.00	94.20
94.40	94.60	94.80	95.00
95.20	95.40	95.60	95.80
96.00	96.20	96.40	96.60
96.80	97.00	97.20	97.40
97.60	97.80	98.00	98.20
98.40	98.60	98.80	99.00
99.20	99.40	99.60	99.80
100.00	100.20	100.40	100.60
100.80	101.00	101.20	101.40
101.60	101.80	102.00	102.20
102.40	102.60	102.80	103.00
103.20	103.40	103.60	103.80
104.00	104.20	104.40	104.60
104.80	105.00	105.20	105.40
105.60	105.80	106.00	106.20
106.40	106.60	106.80	107.00
107.20	107.40	107.60	107.80
108.00	108.20	108.40	108.60
108.80	109.00	109.20	109.40
109.60	109.80	110.00	109.80

12

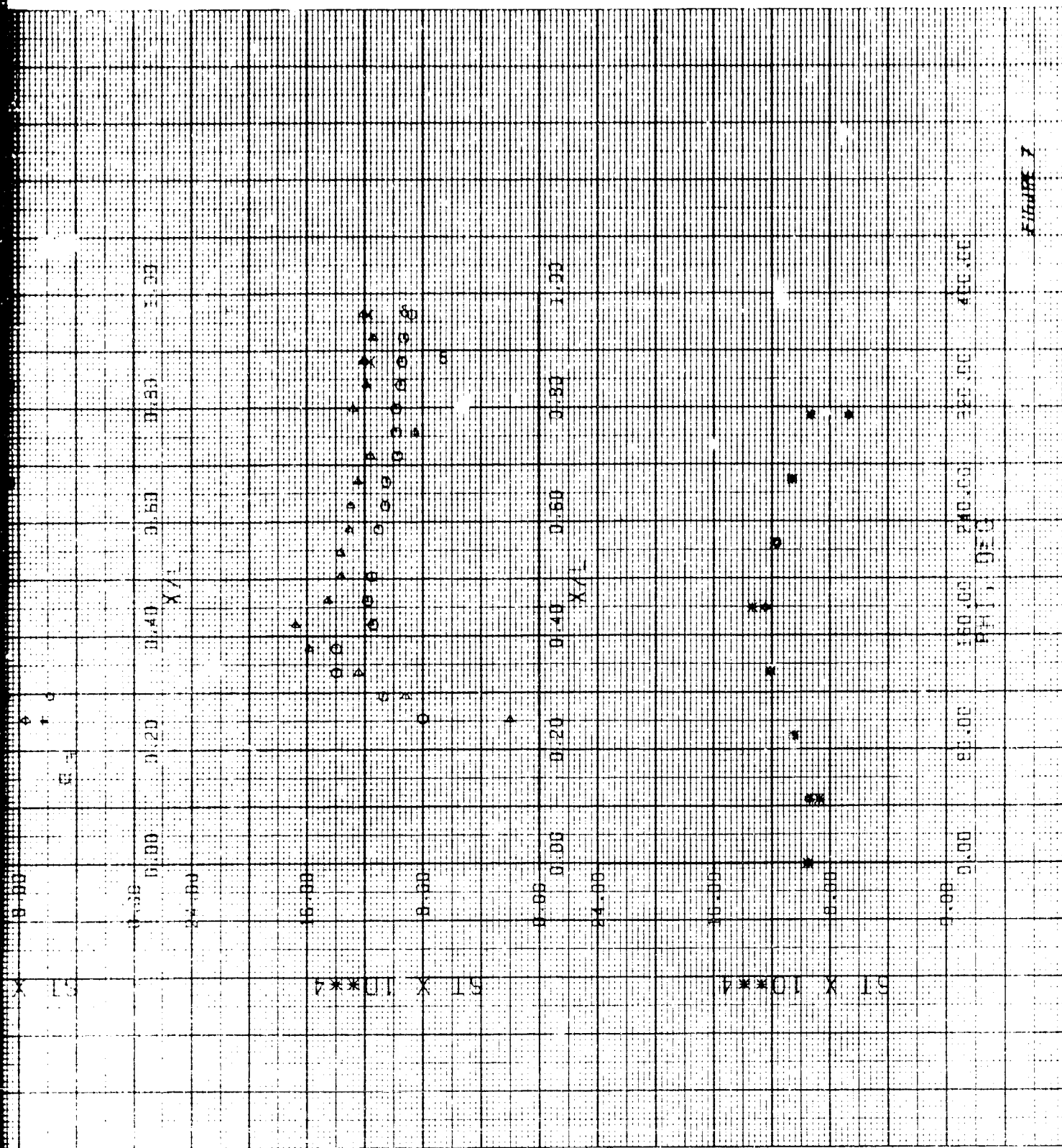


FIGURE 7

Figure 7

7/16/69

MEDICAL INC. - ANNUL AT 5' TENNESSEE  
VON HARMAN GAS DYNAMICS FACILITY  
80 INCH HYPERSONIC TUNNEL #  
VT0947-000

Table with columns: GROUP, COMP ID, ALPHA-M, BETA-M, ALPHA-S, MACH NO, DATE, MACH NO, PO, TO, TIME. It contains a large grid of numerical data points for various test runs.

Vertical grid on the right side of the page, possibly a secondary data table or a reference grid, with multiple columns and rows.

GE	ASYMMETRIC	TRIM	TEST
	HEBI	TRANSFER	PHASE
GROUP B		PHI X/L	SYM PHI X/L
MACH NO. 18		0	225
RE NO. 13179 X 10**4		45	270
CANFIC		90	315
ELPHA P		125	-
TIME 0 SEC		180	-
32-00			
24-00			
16-00			
8-00			
0-00			
32-00		3.50	3.50
24-00		3.40	3.50
16-00			
8-00			
0-00			
32-00			
24-00			
16-00			
8-00			
0-00			

2

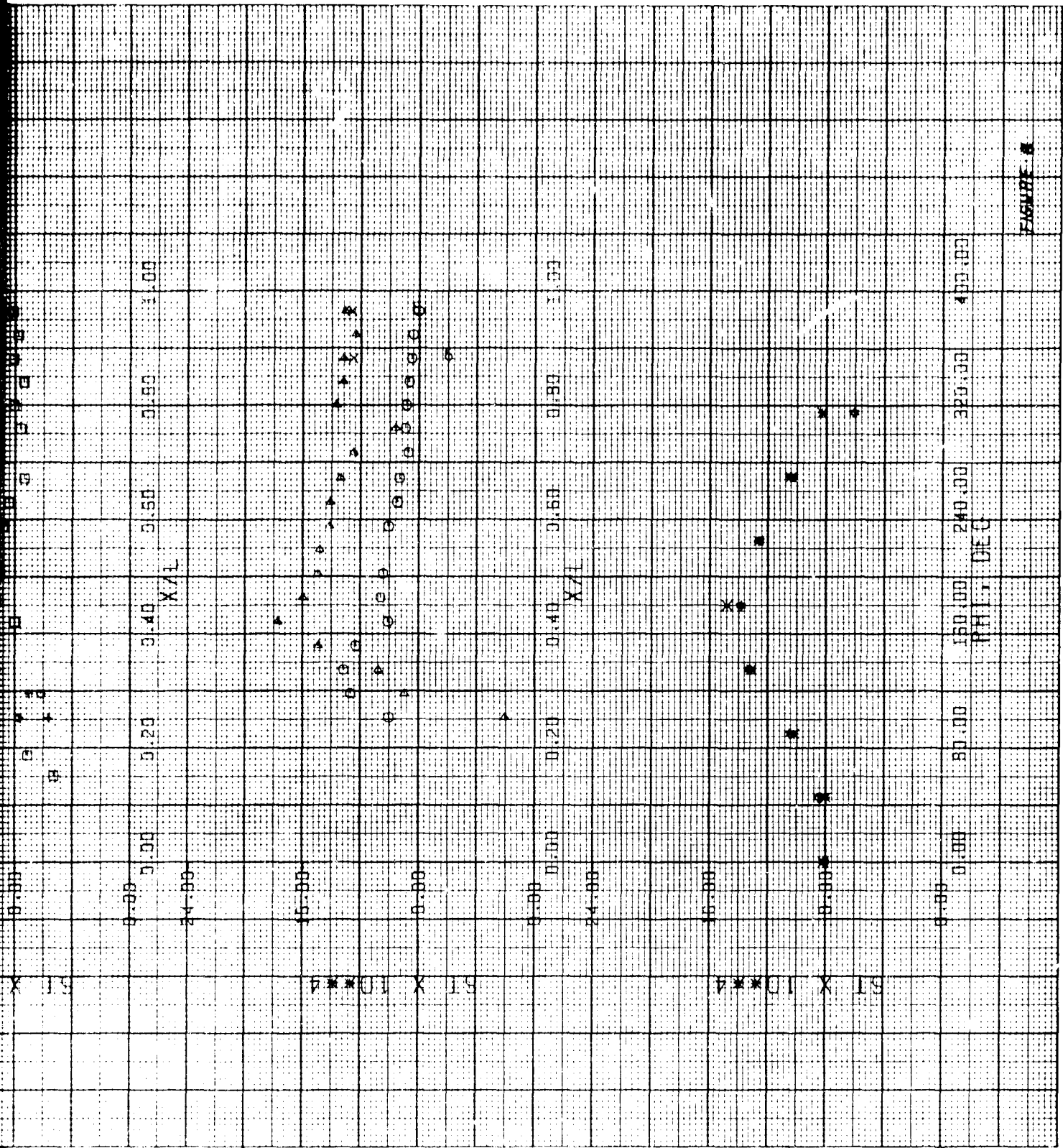


Figure 8

7/16/69

MEDICARD, INC. 1 ANNOLD APTS TENNESSEE  
VON KARMAN GAS DYNAMICS FACILITY  
30 INCH HYPERSONIC TUNNEL 8

V10967-000  
GROUP 4  
COMP ID I ALPHA-A 6.88 BETA-H 6 ALPHA-A 4.35 ROLL 8 ALP-P 8 DATE 716  
T-IMP P-IMP Q-IMP V-IMP RHO-IMP RU-IMP  
9.68152E 01 0.81187E-02 3.94772E 00 3.85747E 03 7.63479E-05 7.74626E-08  
MACH NO PO TO TIME  
8.00 866.29 1336.33 .68  
REF ID A670  
3.77761E 00 3.46502E 00 3.27696E 02

Table with columns: PC NO, R/L, PHI, TEMP, SLOPE, Q-SLOW, HT, HM, BT. Rows 1-100. Data includes numerical values for various parameters across different PC numbers.

GE SYMMETRIC TRIM TEST  
HFBI TRANSFER PHASE  
GROUP 4

GROUP	GE	ASYMMETRIC	TRIM	TEST	PHI	PHI	PHI	PHI	PHI	PHI
MACH NO	HEAT	TRANSFER	PHASE	PHASE	PHASE	PHASE	PHASE	PHASE	PHASE	PHASE
4										
3.78										
10**6										
4										
0										
32.00										
24.00										
16.00										
8.00										
0.00										
32.00										
24.00										
16.00										
8.00										
0.00										

12

SI X 10\*\*6

4

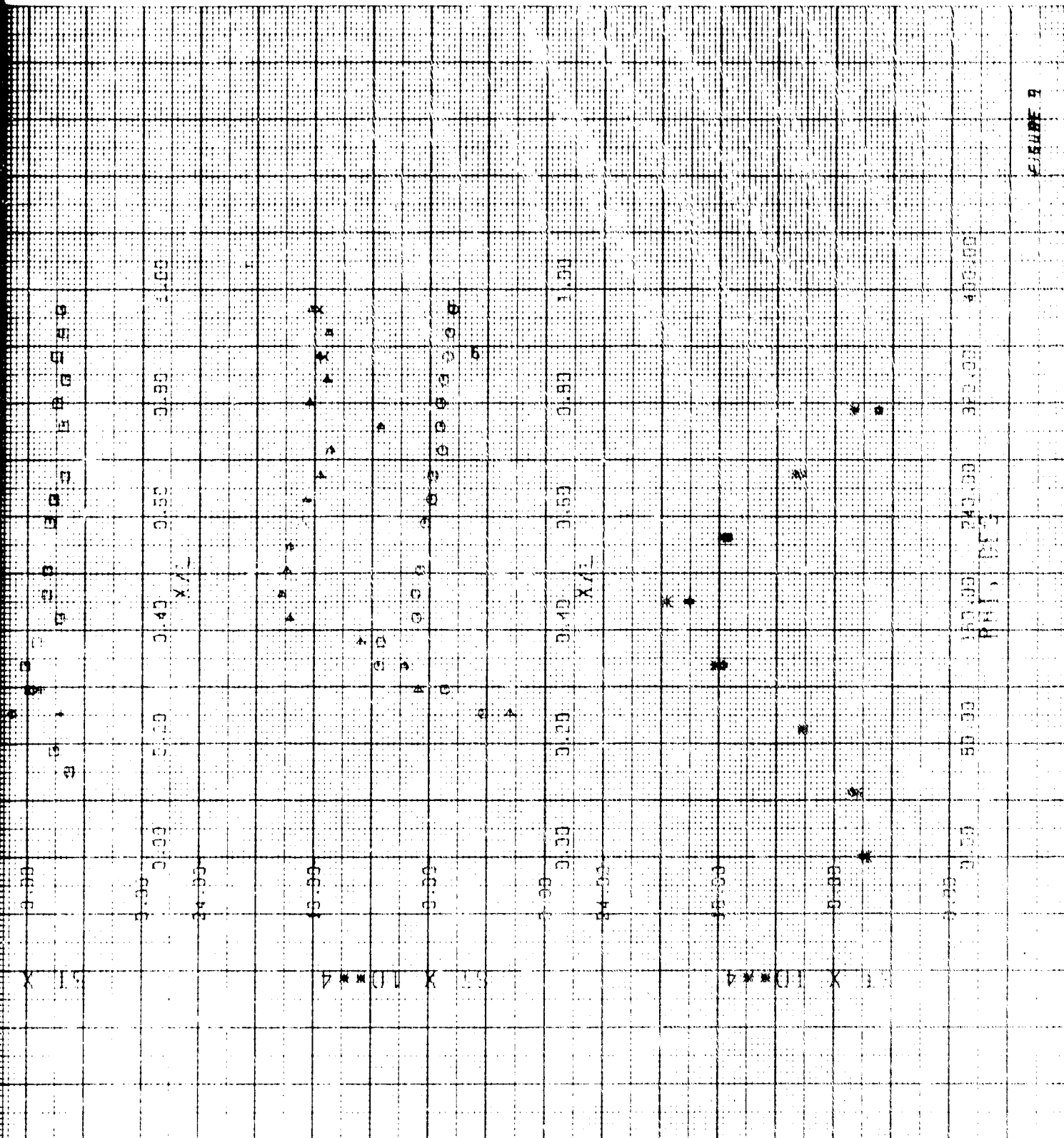


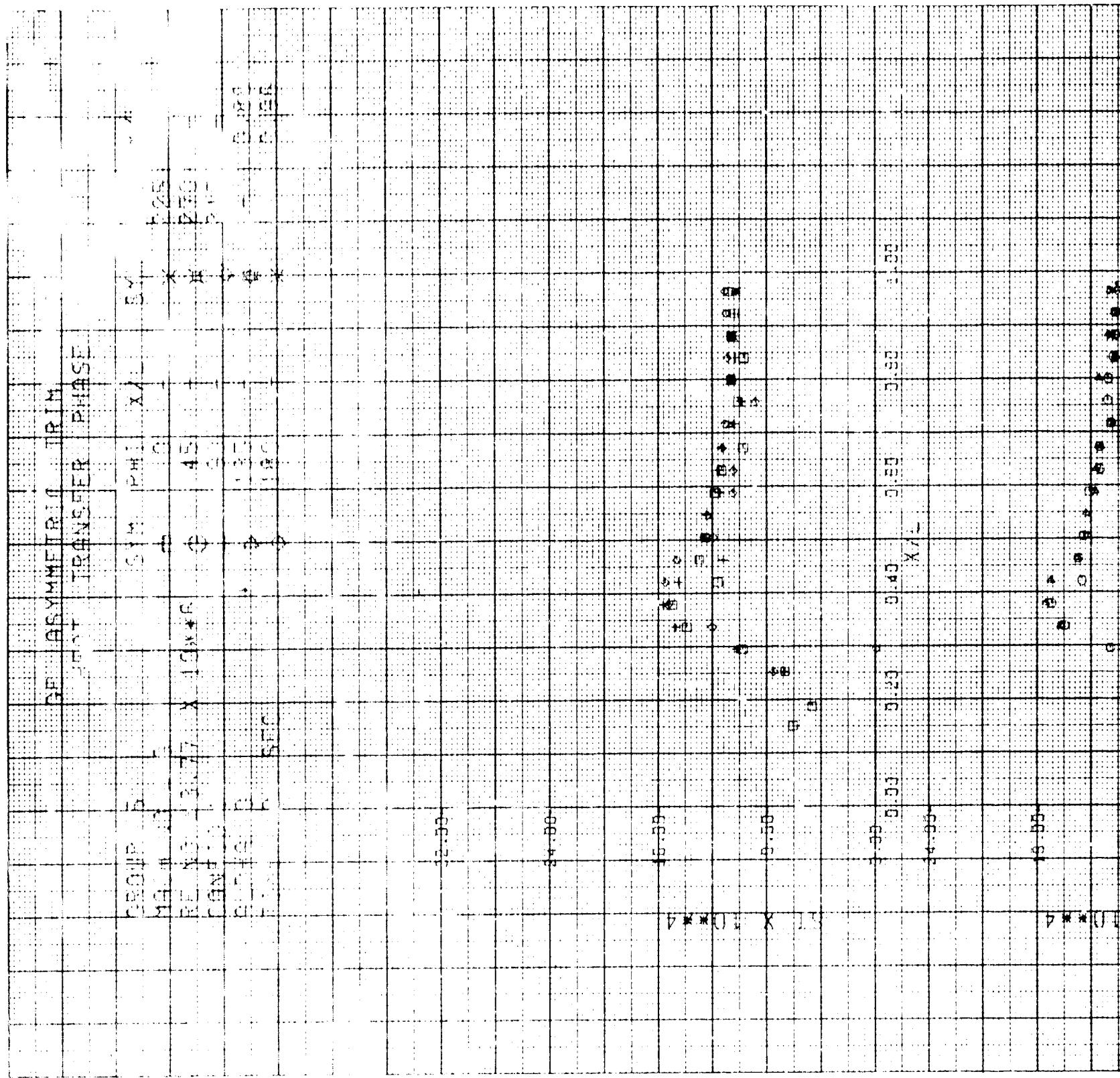
Figure 9

7/12/70

MECONO, INC. - ANNOLD AFS - TENNESSEE  
 Von KARMAN GAS DYNAMICS FACILITY  
 50 INCH HYPERSONIC TUNNEL #

V10967-000  
 ALPHA-4  
 MULL  
 ALP-4 DATE 174  
 RACH NO 0-00  
 PU 886.49  
 TG 1337.51  
 TIME .80  
 RE/PT  
 MU=147  
 MU=118-03  
 3-77307E 04  
 3-65940E 04  
 3-27501E 02

NO	IMP	ALT.	PHASE	TEMP	CHARGE	Q-STON	HT	HM	HU	ST	
1	9500	0		812.51	7.4173E 00	1.1893E 00	1.4417E-03	3.0021E-03	1.2278E 02	0.1243E-04	
2	9500	0		808.20	7.3119E 00	7.2261E-01	1.1099E-03	4.4461E-03	1.2192E 02	4.7163E-04	
3	9500	0		812.84	7.4184E 00	1.1854E 00	1.5804E-03	6.1608E-03	1.2281E 02	6.7143E-04	
4	9500	0		812.84	7.4184E 00	1.1854E 00	1.2020E-03	4.3202E-03	1.2542E 02	9.4513E-04	
5	9500	0		812.84	7.4184E 00	1.1854E 00	2.6611E-03	1.2406E-03	1.2404E 02	1.3967E-03	
6	9500	0		812.84	7.4184E 00	1.1854E 00	2.4447E-03	1.5370E-03	1.4220E-02	1.5020E-03	
7	9500	0		812.84	7.4184E 00	1.1854E 00	2.2215E-03	1.7028E-03	1.1048E-02	1.2725E 02	1.1883E-03
8	9500	0		812.84	7.4184E 00	1.1854E 00	2.4701E-03	1.0184E-03	1.2374E-02	1.2934E 02	1.3062E-03
9	9500	0		812.84	7.4184E 00	1.1854E 00	2.3030E-03	2.4700E-03	1.1938E-02	1.2635E 02	1.2602E-03
10	9500	0		812.84	7.4184E 00	1.1854E 00	1.1933E-03	3.8975E-03	1.3770E-02	1.2689E 02	3.7764E-04
11	9500	0		812.84	7.4184E 00	1.1854E 00	2.2145E-03	2.4040E-03	1.1272E-02	1.2910E 02	1.1888E-03
12	9500	0		812.84	7.4184E 00	1.1854E 00	2.0079E-03	1.0091E-02	1.2904E 02	1.1401E-03	
13	9500	0		812.84	7.4184E 00	1.1854E 00	2.3150E-03	3.3014E-03	1.3016E 02	0.8182E-04	
14	9500	0		812.84	7.4184E 00	1.1854E 00	1.9318E-03	2.7549E-03	1.3082E 02	1.1314E-04	
15	9500	0		812.84	7.4184E 00	1.1854E 00	1.4936E-03	2.3955E-03	1.3117E 02	1.0157E-03	
16	9500	0		812.84	7.4184E 00	1.1854E 00	1.9929E-03	1.0134E-02	1.3134E 02	1.0609E-03	
17	9500	0		812.84	7.4184E 00	1.1854E 00	1.8208E-03	2.3039E-03	1.3171E 02	9.7895E-03	
18	9500	0		812.84	7.4184E 00	1.1854E 00	2.9244E-03	3.0134E-02	1.3200E 02	1.0609E-03	
19	9500	0		812.84	7.4184E 00	1.1854E 00	1.5484E-03	1.0319E-02	1.3205E 02	1.0609E-03	
20	9500	0		812.84	7.4184E 00	1.1854E 00	1.8686E-03	7.5055E-03	1.3226E 02	1.0609E-03	
21	9500	0		812.84	7.4184E 00	1.1854E 00	2.6611E-03	1.0127E-02	1.3245E 02	1.0609E-03	
22	9500	0		812.84	7.4184E 00	1.1854E 00	2.5377E-03	1.0319E-02	1.3245E 02	1.0609E-03	
23	9500	0		812.84	7.4184E 00	1.1854E 00	1.5484E-03	1.0319E-02	1.3245E 02	1.0609E-03	
24	9500	0		812.84	7.4184E 00	1.1854E 00	2.5377E-03	1.0319E-02	1.3245E 02	1.0609E-03	
25	9500	0		812.84	7.4184E 00	1.1854E 00	1.8686E-03	1.0319E-02	1.3245E 02	1.0609E-03	
26	9500	0		812.84	7.4184E 00	1.1854E 00	1.5484E-03	1.0319E-02	1.3245E 02	1.0609E-03	
27	9500	0		812.84	7.4184E 00	1.1854E 00	1.8686E-03	1.0319E-02	1.3245E 02	1.0609E-03	
28	9500	0		812.84	7.4184E 00	1.1854E 00	1.5484E-03	1.0319E-02	1.3245E 02	1.0609E-03	
29	9500	0		812.84	7.4184E 00	1.1854E 00	1.8686E-03	1.0319E-02	1.3245E 02	1.0609E-03	
30	9500	0		812.84	7.4184E 00	1.1854E 00	1.5484E-03	1.0319E-02	1.3245E 02	1.0609E-03	
31	9500	0		812.84	7.4184E 00	1.1854E 00	1.8686E-03	1.0319E-02	1.3245E 02	1.0609E-03	
32	9500	0		812.84	7.4184E 00	1.1854E 00	1.5484E-03	1.0319E-02	1.3245E 02	1.0609E-03	
33	9500	0		812.84	7.4184E 00	1.1854E 00	1.8686E-03	1.0319E-02	1.3245E 02	1.0609E-03	
34	9500	0		812.84	7.4184E 00	1.1854E 00	1.5484E-03	1.0319E-02	1.3245E 02	1.0609E-03	
35	9500	0		812.84	7.4184E 00	1.1854E 00	1.8686E-03	1.0319E-02	1.3245E 02	1.0609E-03	
36	9500	0		812.84	7.4184E 00	1.1854E 00	1.5484E-03	1.0319E-02	1.3245E 02	1.0609E-03	
37	9500	0		812.84	7.4184E 00	1.1854E 00	1.8686E-03	1.0319E-02	1.3245E 02	1.0609E-03	
38	9500	0		812.84	7.4184E 00	1.1854E 00	1.5484E-03	1.0319E-02	1.3245E 02	1.0609E-03	
39	9500	0		812.84	7.4184E 00	1.1854E 00	1.8686E-03	1.0319E-02	1.3245E 02	1.0609E-03	
40	9500	0		812.84	7.4184E 00	1.1854E 00	1.5484E-03	1.0319E-02	1.3245E 02	1.0609E-03	
41	9500	0		812.84	7.4184E 00	1.1854E 00	1.8686E-03	1.0319E-02	1.3245E 02	1.0609E-03	
42	9500	0		812.84	7.4184E 00	1.1854E 00	1.5484E-03	1.0319E-02	1.3245E 02	1.0609E-03	
43	9500	0		812.84	7.4184E 00	1.1854E 00	1.8686E-03	1.0319E-02	1.3245E 02	1.0609E-03	
44	9500	0		812.84	7.4184E 00	1.1854E 00	1.5484E-03	1.0319E-02	1.3245E 02	1.0609E-03	
45	9500	0		812.84	7.4184E 00	1.1854E 00	1.8686E-03	1.0319E-02	1.3245E 02	1.0609E-03	
46	9500	0		812.84	7.4184E 00	1.1854E 00	1.5484E-03	1.0319E-02	1.3245E 02	1.0609E-03	
47	9500	0		812.84	7.4184E 00	1.1854E 00	1.8686E-03	1.0319E-02	1.3245E 02	1.0609E-03	
48	9500	0		812.84	7.4184E 00	1.1854E 00	1.5484E-03	1.0319E-02	1.3245E 02	1.0609E-03	
49	9500	0		812.84	7.4184E 00	1.1854E 00	1.8686E-03	1.0319E-02	1.3245E 02	1.0609E-03	
50	9500	0		812.84	7.4184E 00	1.1854E 00	1.5484E-03	1.0319E-02	1.3245E 02	1.0609E-03	
51	9500	0		812.84	7.4184E 00	1.1854E 00	1.8686E-03	1.0319E-02	1.3245E 02	1.0609E-03	
52	9500	0		812.84	7.4184E 00	1.1854E 00	1.5484E-03	1.0319E-02	1.3245E 02	1.0609E-03	
53	9500	0		812.84	7.4184E 00	1.1854E 00	1.8686E-03	1.0319E-02	1.3245E 02	1.0609E-03	
54	9500	0		812.84	7.4184E 00	1.1854E 00	1.5484E-03	1.0319E-02	1.3245E 02	1.0609E-03	
55	9500	0		812.84	7.4184E 00	1.1854E 00	1.8686E-03	1.0319E-02	1.3245E 02	1.0609E-03	
56	9500	0		812.84	7.4184E 00	1.1854E 00	1.5484E-03	1.0319E-02	1.3245E 02	1.0609E-03	
57	9500	0		812.84	7.4184E 00	1.1854E 00	1.8686E-03	1.0319E-02	1.3245E 02	1.0609E-03	
58	9500	0		812.84	7.4184E 00	1.1854E 00	1.5484E-03	1.0319E-02	1.3245E 02	1.0609E-03	
59	9500	0		812.84	7.4184E 00	1.1854E 00	1.8686E-03	1.0319E-02	1.3245E 02	1.0609E-03	
60	9500	0		812.84	7.4184E 00	1.1854E 00	1.5484E-03	1.0319E-02	1.3245E 02	1.0609E-03	
61	9500	0		812.84	7.4184E 00	1.1854E 00	1.8686E-03	1.0319E-02	1.3245E 02	1.0609E-03	
62	9500	0		812.84	7.4184E 00	1.1854E 00	1.5484E-03	1.0319E-02	1.3245E 02	1.0609E-03	
63	9500	0		812.84	7.4184E 00	1.1854E 00	1.8686E-03	1.0319E-02	1.3245E 02	1.0609E-03	
64	9500	0		812.84	7.4184E 00	1.1854E 00	1.5484E-03	1.0319E-02	1.3245E 02	1.0609E-03	
65	9500	0		812.84	7.4184E 00	1.1854E 00	1.8686E-03	1.0319E-02	1.3245E 02	1.0609E-03	
66	9500	0		812.84	7.4184E 00	1.1854E 00	1.5484E-03	1.0319E-02	1.3245E 02	1.0609E-03	
67	9500	0		812.84	7.4184E 00	1.1854E 00	1.8686E-03	1.0319E-02	1.3245E 02	1.0609E-03	
68	9500	0		812.84	7.4184E 00	1.1854E 00	1.5484E-03	1.0319E-02	1.3245E 02	1.0609E-03	
69	9500	0		812.84	7.4184E 00	1.1854E 00	1.8686E-03	1.0319E-02	1.3245E 02	1.0609E-03	
70	9500	0		812.84	7.4184E 00	1.1854E 00	1.5484E-03	1.0319E-02	1.3245E 02	1.0609E-03	
71	9500	0		812.84	7.4184E 00	1.1854E 00	1.8686E-03	1.0319E-02	1.3245E 02	1.0609E-03	
72	9500	0		812.84	7.4184E 00	1.1854E 00	1.5484E-03	1.0319E-02	1.3245E 02	1.0609E-03	
73	9500	0		812.84	7.4184E 00	1.1854E 00	1.8686E-03	1.0319E-02	1.3245E 02	1.0609E-03	
74	9500	0		812.84	7.4184E 00	1.1854E 00	1.5484E-03	1.0319E-02	1.3245E 02	1.0609E-03	
75	9500	0		812.84	7.4184E 00	1.1854E 00	1.8686E-03	1.0319E-02	1.3245E 02	1.0609E-03	
76	9500	0		812.84	7.4184E 00	1.1854E 00	1.5484E-03	1.0319E-02	1.3245E 02	1.0609E-03	
77	9500	0		812.84	7.4184E 00	1.1854E 00	1.8686E-03	1.0319E-02	1.3245E 02	1.0609E-03	
78	9500	0		812.84	7.4184E 00	1.1854E 00	1.5484E-03	1.0319E-02	1.3245E 02	1.0609E-03	
79	9500	0		812.84	7.4184E 00	1.1854E 00	1.8686E-03	1.0319E-02	1.3245E 02	1.0609E-03	
80	9500	0		812.84	7.4184E 00	1.1854E 00	1.5484E-03	1.0319E-02	1.3245E 02	1.0609E-03	
81	9500	0		812.84	7.4184E 00	1.1854E 00	1.8686E-03	1.0319E-02	1.3245E 02	1.0609E-03	
82	9500	0		812.84	7.4184E 00	1.1854E 00	1.5484E-03	1.0319E-02	1.3245E 02	1.0609E-03	
83	9500	0		812.84	7.4184E 00	1.1854E 00	1.8686E-03	1.0319E-02	1.3245E 02	1.0609E-03	
84	9500	0		812.84	7.4184E 00	1.1854E 00	1.5484E-03	1.0319E-02	1.3245E 02	1.0609E-03	
85	9500	0		812.84	7.4184E 00	1.1854E 00	1.8686E-03	1.0319E-02	1.3245E 02	1.0609E-03	
86	9500	0		812.84	7.4184E 00	1.1854E 00	1.5484E-03	1.0319E-02	1.3245E 02	1.0609E-03	
87	9500	0		812.84	7.4184E 00	1.1854E 00	1.8686E-03	1.0319E-02	1.3245E 02	1.0609E-03	
88	9500	0		812.84	7.4184E 00	1.1854E 00	1.5484E-03	1.0319E-02	1.3245E 02	1.0609E-03	
89	9500	0		812.84	7.4184E 00	1.1854E 00	1.8686E-03	1.0319E-02	1.3245E 02	1.0609E-03	
90	9500	0		812.84	7.4184E 00	1.1854E 00	1.5484E-03	1.0319E-02	1.3245E 02	1	



2

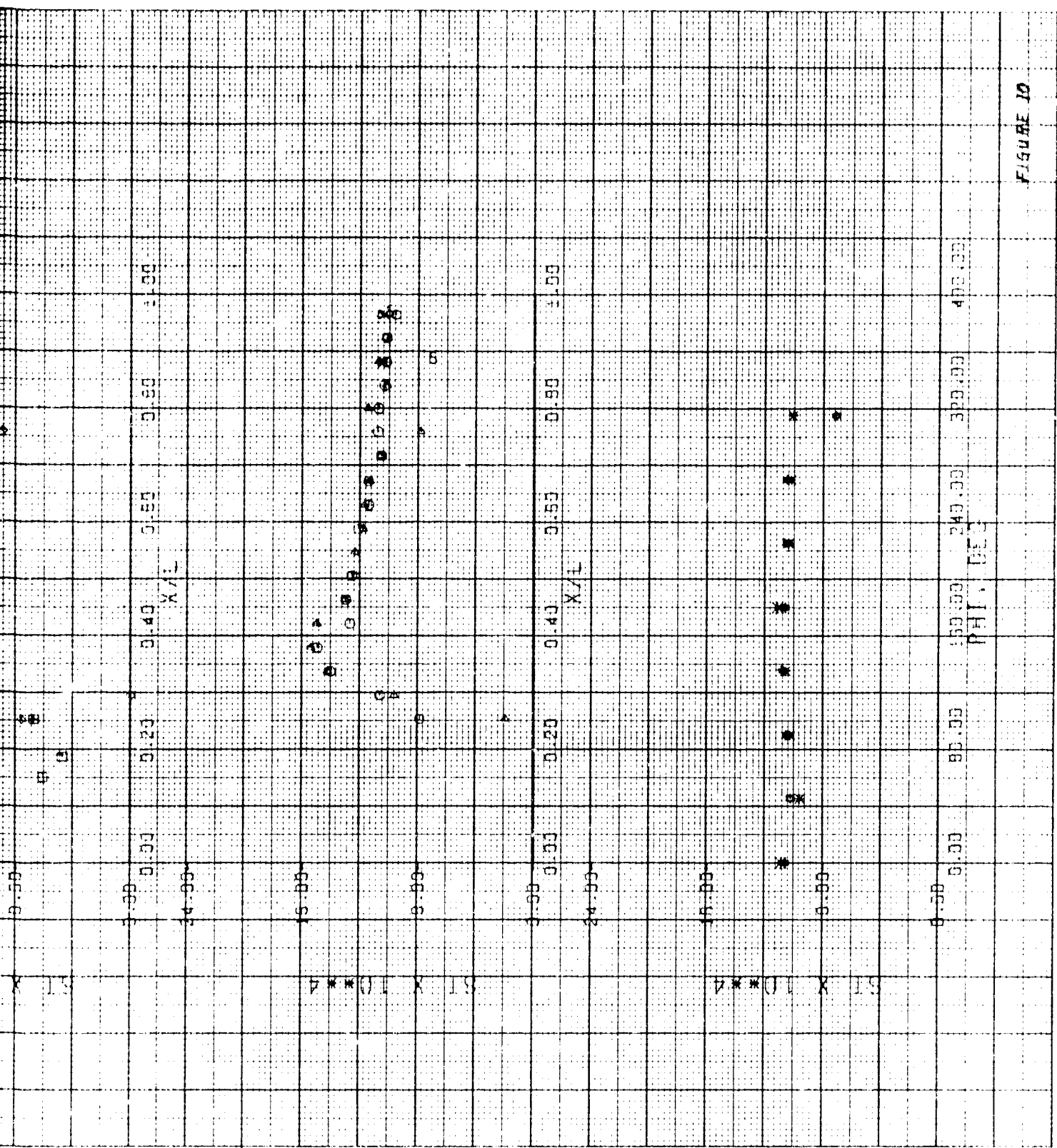


FIGURE 10

Figure 10

3

7/16/69

AMERICAN-INC.) ANNULD APS- TENNESSEE  
WIND HAMMAN GAS DYNAMICS FACILITY  
SO INCM HYPERSONIC TUNNEL #  
WTOPA-M00

Table with columns: GROUP, CONFID, ALPHA-N, BETA-N, ALPHA-N, MACH, DATE, MACH NO, PO, TO, TIME. Rows include data for various test conditions and parameters.

ASYMMETRIC TRIM TEST  
HEAT TRANSFER PHASE  
GROUP 1 ISYM PHI X Y Z

2

GROUP	II	AS	SYMMETRIC	TRIM	TEST
MACH NO	1-8	HEAT	TRANSFER	PHASE	
LEN	NO	3.02	X	10**6	
CONFIG	I				
ALPHA	0				
TIME	0	SEC			
32.00					
34.00					
36.00					
38.00					
40.00					
42.00					
44.00					
46.00					
48.00					
50.00					
52.00					
54.00					
56.00					
58.00					
60.00					
62.00					
64.00					
66.00					
68.00					
70.00					
72.00					
74.00					
76.00					
78.00					
80.00					
82.00					
84.00					
86.00					
88.00					
90.00					
92.00					
94.00					
96.00					
98.00					
100.00					

ST X 10\*\*4

10\*\*4

00000000

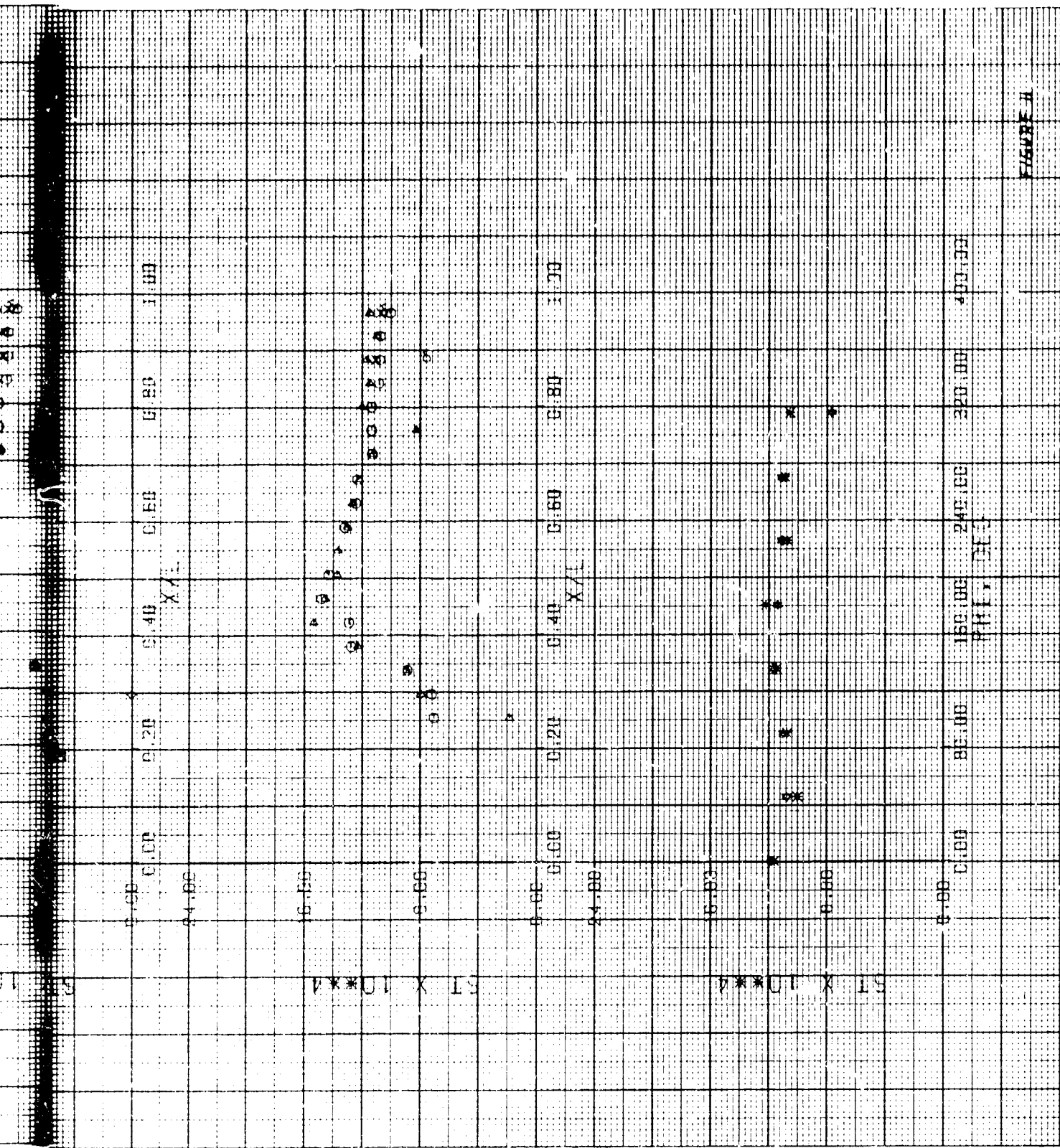


FIGURE 11

Figure 11

7/16/69

AUXILIARY... ARMOURED AIR FORCE TENSIVE...  
MUN... DYNAMIC FACILITY...  
... TOWER...  
... MOOD

Table with columns: GROUP, COMP ID, ALPHA-H, BETA-H, ALPHA-H, ALPHA-H, DATE, MACH NO, PD, TO, TIME. Rows contain numerical data for various components.

GROUP 12  
PHI X11 SYM DEF 331  
HEFT TRANSFER PHASE  
IRIM TEST

GROUP	PHI	PHI	PHI	PHI	PHI	PHI	PHI
12	15	90	135	180	225	270	315
12.00	15.00	90.00	135.00	180.00	225.00	270.00	315.00
24.00							
36.00							
48.00							
60.00							
72.00							
84.00							
96.00							
108.00							
120.00							
132.00							
144.00							
156.00							
168.00							
180.00							
192.00							
204.00							
216.00							
228.00							
240.00							
252.00							
264.00							
276.00							
288.00							
300.00							
312.00							
324.00							
336.00							
348.00							
360.00							

2

ST X 10\*\*4

\*\*4

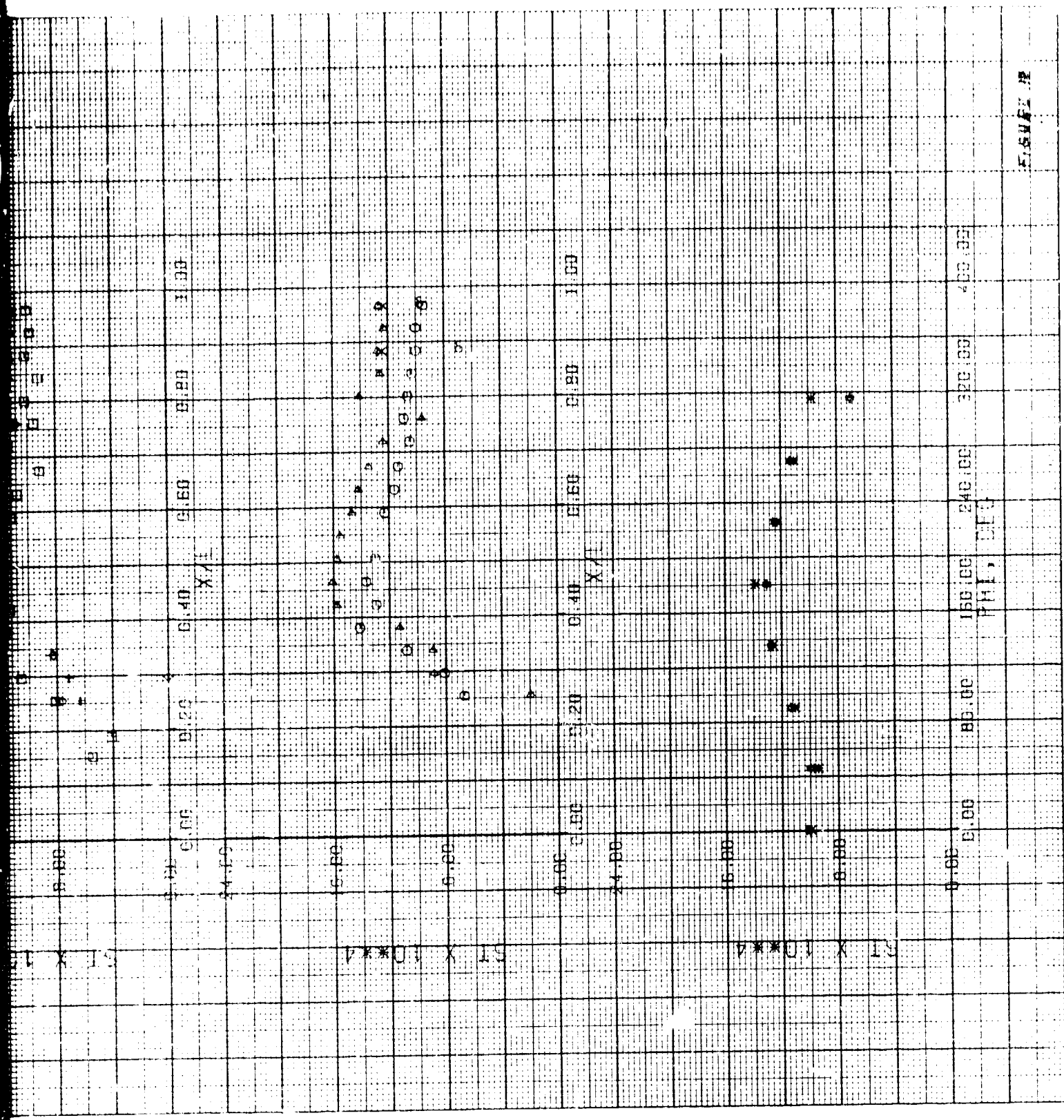
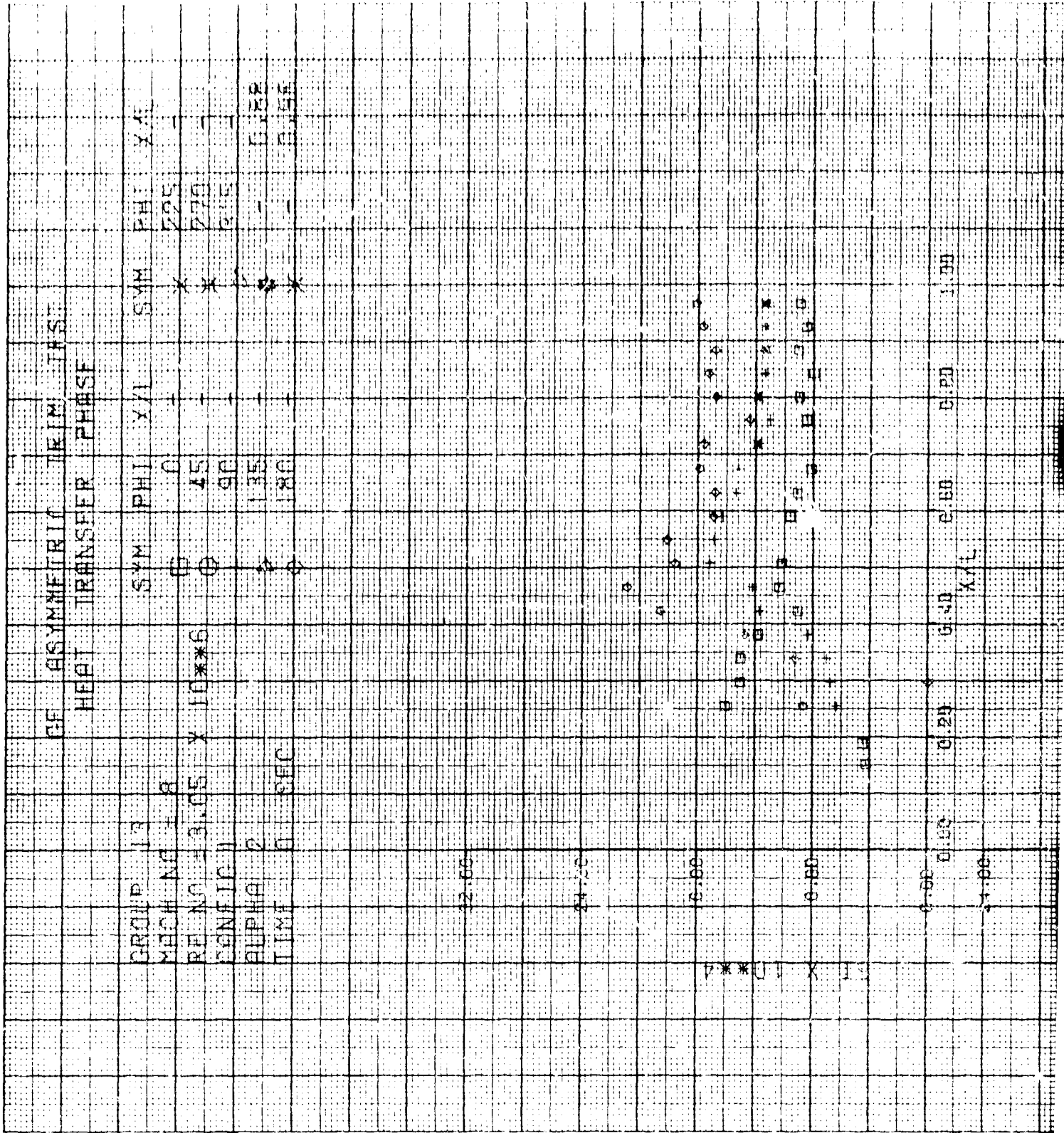


Figure 12

W





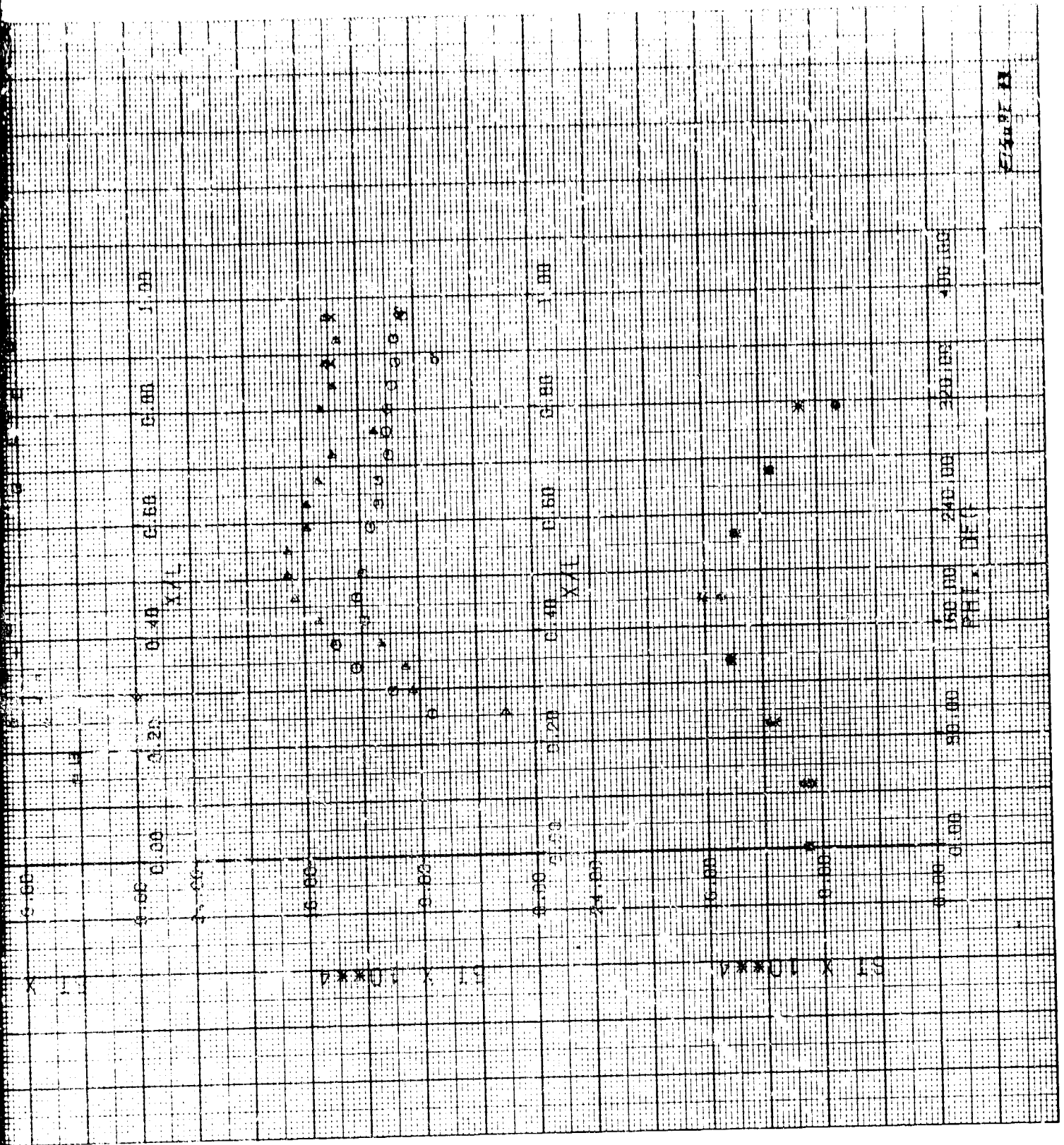


Figure 13



CF ASYMMETRIC TRIM TEST  
HEAT TRANSFER PHASE

GROUP 14  
MACH NO = 8  
REF NO = 3.04 X 10\*\*6  
GEOMETRIC 1  
ALPHA 0  
TIME 0 SEC

SYN PHI X/L SYN PHI X/L  
 □ 0  
 ○ 45  
 △ 90  
 \* 135  
 ◇ 180

PHI X/L  
 225  
 270  
 315  
 360  
 0

32.00

34.00

36.00

38.00

40.00

42.00

44.00

3.10\*\*4

3.00

3.20

3.40

3.60

3.80

4.00

4.20

4.40

4.60

4.80

5.00

5.20

2

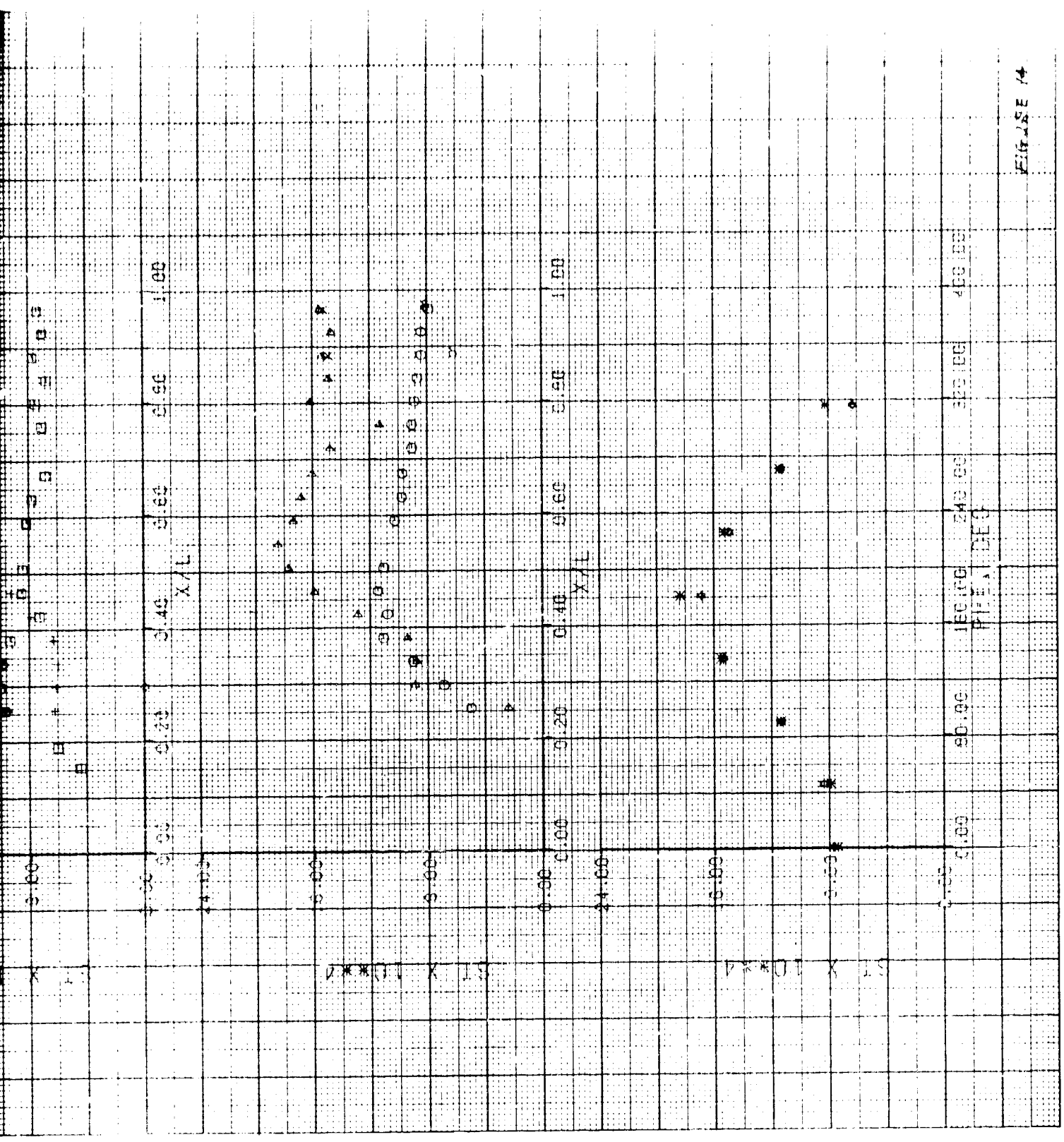
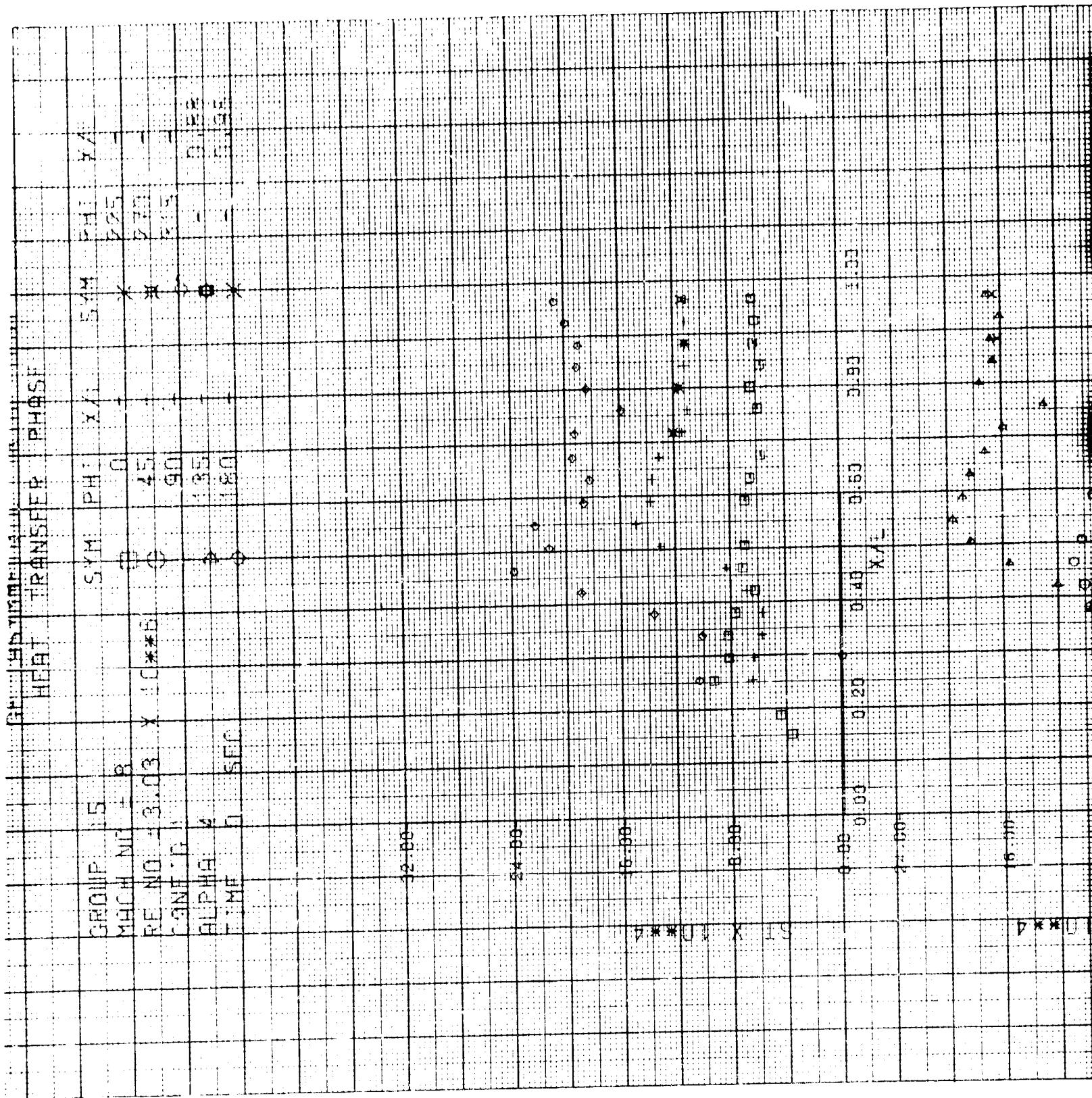


FIGURE 14

Figure 14





GROUP 15  
 MACH NO = 8  
 RE NO = 3.03 X 10\*\*5  
 CONF. 1  
 ALPHA 4  
 TIME 0 SEC

SYM PHI XAI SYM PHI XAI  
 \* \* \* \* \*  
 \* \* \* \* \*  
 \* \* \* \* \*  
 \* \* \* \* \*  
 \* \* \* \* \*

PH I YAI  
 225  
 270  
 315  
 0  
 0

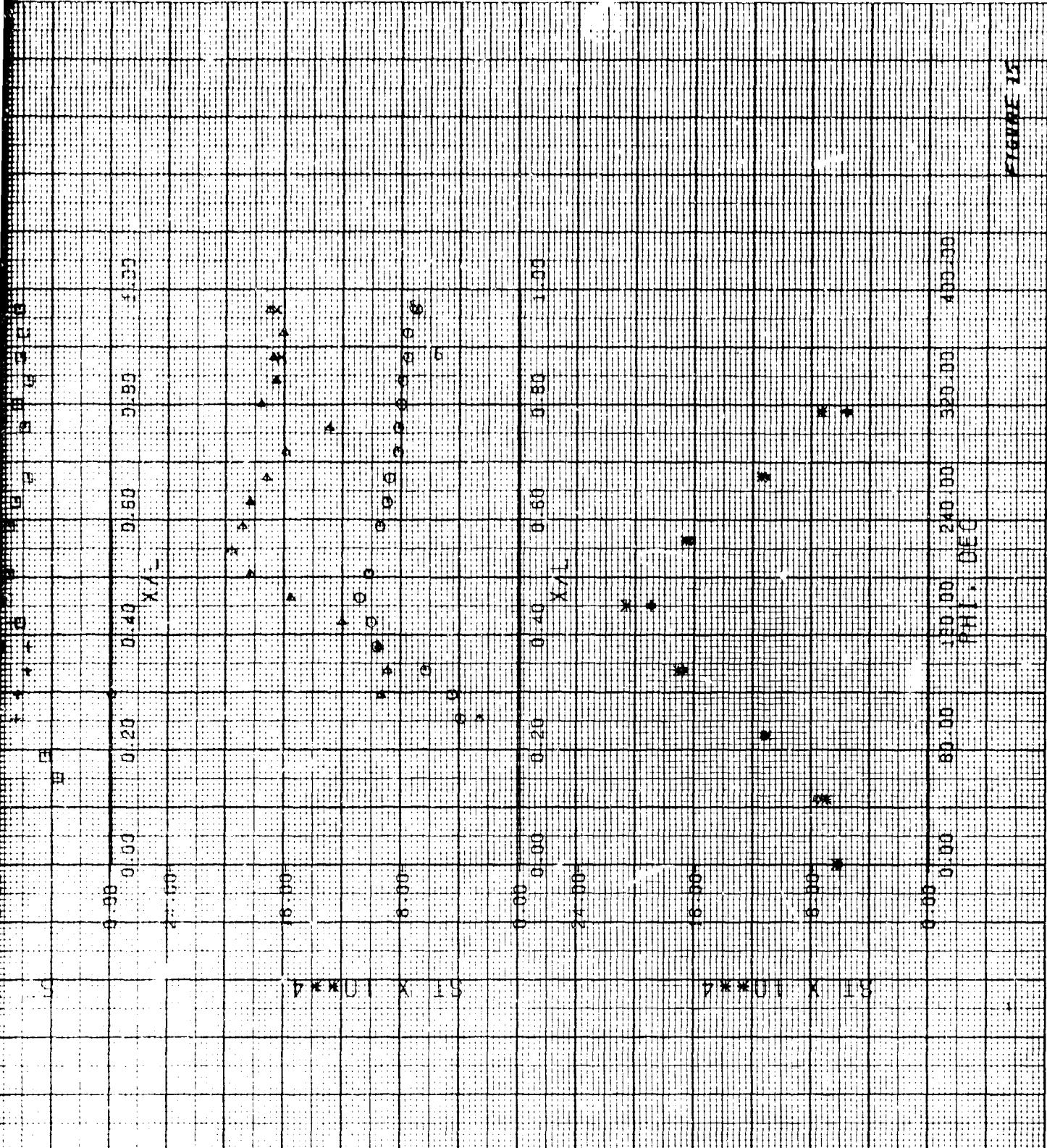


FIGURE 15

Figure 15

3



CE ASYMMETRIC TRIM TEST  
HEAT TRANSFER PHASE

GROUP 16  
MACH NO 1.8  
RE NO = 3.03 X 10\*\*6  
CONFIG 1  
ALPHA 0  
TIME 0 SEC

SYM PHI XXL SYM PHI XL  
\* \* \* \* \*  
\* \* \* \* \*  
\* \* \* \* \*

0.88  
0.96

32-00

34-00

36-00

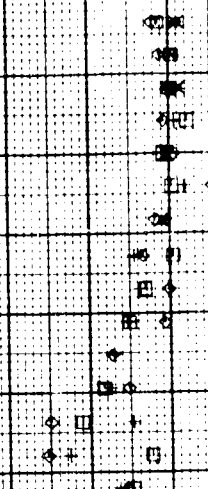
38-00

39-00

40-00

41-00

\*\*\*UL X



17

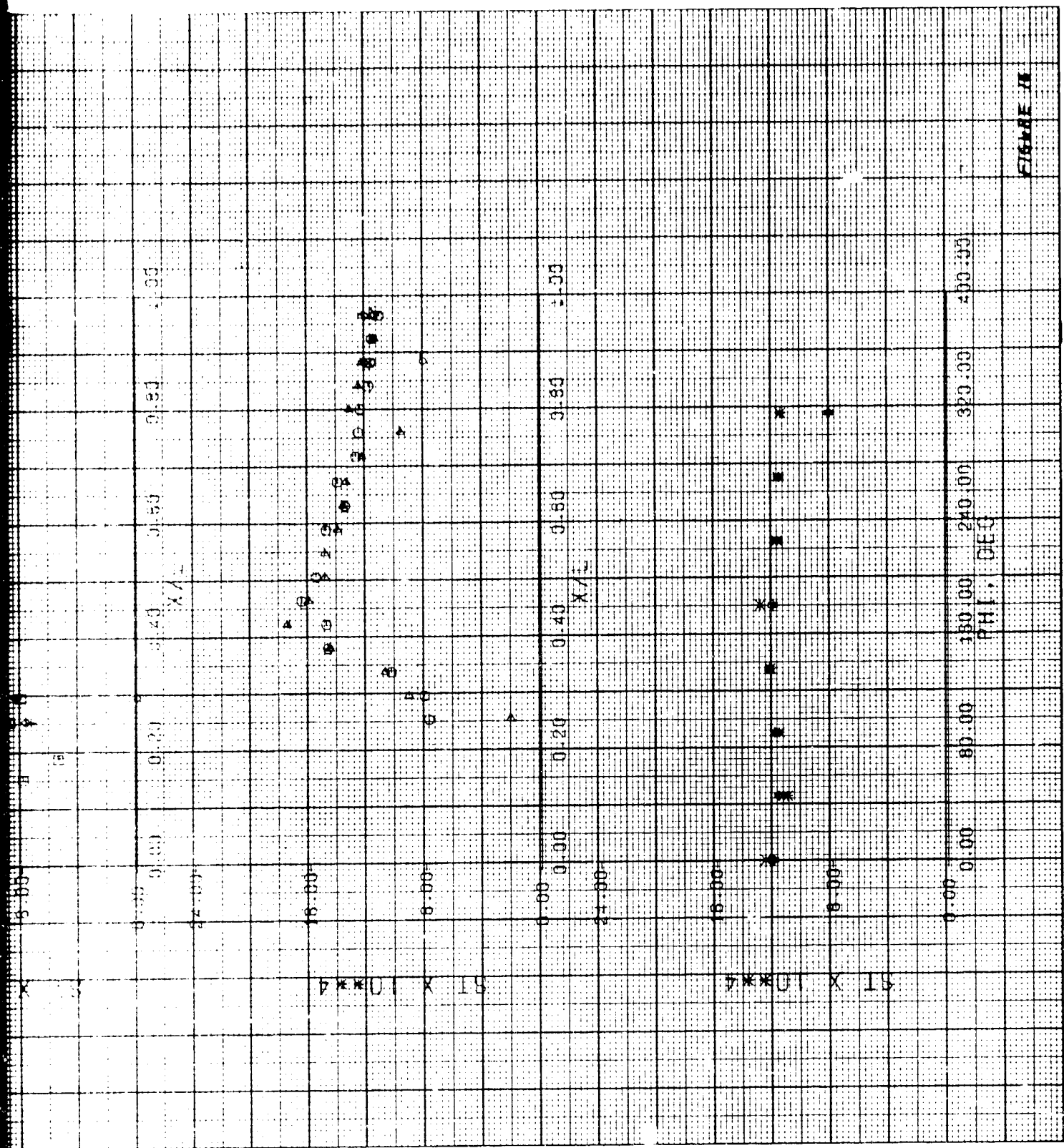


Figure 16

5





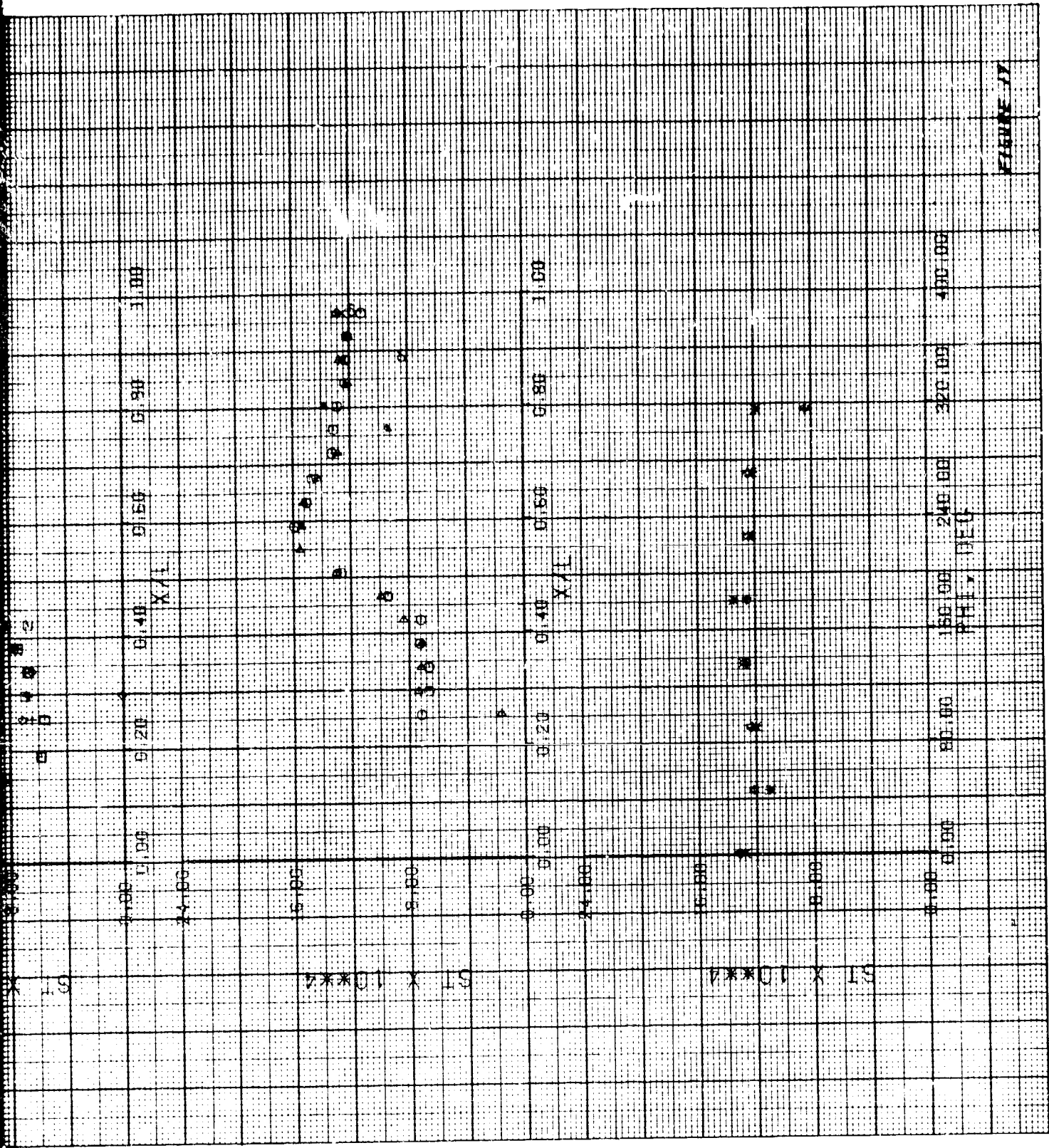


Figure 17

7/14/69

MECHANICAL INC. ANNULD AFS. TENNESSEE  
WON HANMAN GAS DYNAMICS FACILITY  
50 INCH HYPERSONIC TUNNEL

VT047-B00

GROUP	COMP ID	ALPHA-N	BETA-N	ALPHA-K	MULL	ALPH-U	DATE	MACH NO	PHI	TO	TIME
1	1	1.18	8	1.18		U	714	7.09	68.87	1291.97	
U-1W	U-1W	U-1W	U-1W	U-1W	RHO-1W	U-1W	RZ/RT	RZ/RT	NO	NO	
9.38194E-01	4.84171E-02	2.14169E-00	3.37457E-11	4.32894E-05	1.37494E-08	2.17312E-06	1.99314E-06	3.16165E-02			
PC NO	K/L	PHI	T-HP	SI-OPH	U-STO6	HT	HM	HM	ST		
1	1390	0	471.19	7.2843E-08	7.4939E-01	6.4394E-10	3.9164E-03	1.8846E-02	6.4877E-04		
2	1647	0	455.30	2.2286E-00	4.8944E-01	5.4849E-09	2.3672E-03	1.6843E-02	4.4022E-04		
3	2524	0	468.57	2.4612E-00	1.7772E-01	4.1137E-09	1.8874E-03	1.1089E-02	6.4407E-04		
4	2943	0	468.84	2.1765E-00	6.8694E-01	4.3464E-09	1.1722E-03	1.1261E-02	6.3497E-04		
5	3363	0	475.64	1.9420E-00	9.6026E-01	1.1760E-03	4.7325E-03	1.1489E-02	4.9047E-04		
6	3783	0	482.82	1.3647E-00	1.1880E-00	1.4040E-03	5.2418E-03	1.1243E-02	1.1280E-03		
7	4203	0	487.41	1.4093E-00	1.0749E-00	1.7360E-03	5.3976E-03	1.1702E-02	1.0218E-03		
8	4623	0	490.44	5.1382E-00	1.4700E-00	1.4881E-03	1.4044E-03	1.1866E-02	1.1811E-03		
9	5043	0	498.14	5.1359E-00	1.4430E-00	1.4881E-03	1.3020E-03	1.1950E-02	1.3400E-03		
10	5463	0	512.14	5.4156E-02	1.0317E-02	2.1474E-05	6.8543E-05	1.2764E-02	1.6384E-05		
11	5882	0	505.94	4.7240E-00	1.3027E-00	1.0574E-03	6.4640E-03	1.2182E-02	1.2864E-03		
12	6302	0	518.44	4.4499E-00	1.2285E-00	1.5729E-03	6.3464E-03	1.2280E-02	1.2315E-03		
13	6722	0	529.94	4.2612E-00	1.0181E-00	1.3203E-03	5.3242E-03	1.2480E-02	1.0089E-03		
14	7141	315.0	529.94	4.2612E-00	1.0181E-00	1.3203E-03	5.3242E-03	1.2480E-02	1.0089E-03		
15	7561	0	521.72	4.3322E-00	1.0861E-00	1.0242E-03	4.1981E-03	1.2711E-02	1.0732E-04		
16	7980	0	521.41	4.3322E-00	1.1327E-00	1.1841E-03	5.2400E-03	1.2580E-02	1.0786E-03		
17	8400	0	521.41	4.3322E-00	1.1327E-00	1.1841E-03	5.2400E-03	1.2580E-02	1.0786E-03		
18	8819	0	521.41	4.3322E-00	1.1327E-00	1.1841E-03	5.2400E-03	1.2580E-02	1.0786E-03		
19	9238	0	527.39	4.3588E-00	1.1275E-00	1.4702E-03	5.2277E-03	1.2548E-02	1.1222E-03		
20	9658	0	532.04	4.4109E-00	1.0861E-00	1.3948E-03	5.0226E-03	1.2580E-02	1.0644E-03		
21	10077	0	532.04	4.4109E-00	1.0861E-00	1.3948E-03	5.0226E-03	1.2580E-02	1.0644E-03		
22	10496	45.0	458.79	2.3538E-00	6.9580E-01	4.3121E-09	1.3742E-03	1.1026E-02	6.3472E-04		
23	10915	45.0	465.74	2.1440E-00	6.8765E-01	4.3232E-09	1.3088E-03	1.1141E-02	6.3734E-04		
24	11334	45.0	472.11	2.2440E-00	6.7333E-01	4.2126E-09	1.3240E-03	1.1140E-02	6.2807E-04		
25	11753	45.0	478.41	2.4078E-00	6.1496E-01	4.4800E-09	1.4844E-03	1.1494E-02	1.5899E-04		
26	12172	45.0	484.41	2.4078E-00	6.1496E-01	4.4800E-09	1.4844E-03	1.1494E-02	1.5899E-04		
27	12591	45.0	491.91	4.2140E-00	1.1830E-00	1.4799E-03	2.9757E-03	1.1800E-02	1.1313E-03		
28	13010	45.0	498.47	4.4066E-00	1.4357E-00	1.4071E-03	1.2247E-03	1.1940E-02	1.3014E-03		
29	13429	315.0	537.13	4.7393E-00	1.3471E-00	1.5099E-03	6.0819E-03	1.2822E-02	1.1517E-03		
30	13848	45.0	504.32	5.4543E-00	3.8422E-00	1.4907E-03	1.7704E-03	1.2102E-02	1.4593E-03		
31	14267	45.0	509.72	4.1917E-00	1.1954E-00	1.7838E-03	1.1747E-03	1.2731E-02	1.3026E-03		
32	14686	45.0	522.22	4.1635E-00	1.3166E-00	1.7184E-03	6.0770E-03	1.2527E-02	1.3057E-03		
33	15105	45.0	525.43	5.0884E-00	1.1972E-00	1.5623E-03	6.2483E-03	1.2400E-02	1.1424E-03		
34	15524	45.0	528.43	4.4799E-00	1.2343E-00	1.4002E-03	6.4532E-03	1.2488E-02	1.2417E-03		
35	15943	45.0	529.41	4.2966E-00	1.2839E-00	1.5883E-03	6.2979E-03	1.2488E-02	1.1412E-03		
36	16362	45.0	532.72	4.2966E-00	1.1825E-00	1.5112E-03	6.0415E-03	1.2530E-02	1.1536E-03		
37	16781	45.0	529.94	4.2612E-00	1.1227E-00	1.4637E-03	5.4913E-03	1.2492E-02	1.1172E-03		
38	17200	45.0	528.11	4.3397E-00	1.1884E-00	1.4557E-03	5.4527E-03	1.2669E-02	1.1879E-03		
39	17619	45.0	533.41	4.2966E-00	1.4317E-00	1.3088E-03	5.4421E-03	1.2749E-02	1.4878E-03		
40	18038	90.0	463.21	2.4985E-00	7.3621E-01	4.4832E-09	1.3930E-03	1.1130E-02	6.4032E-04		
41	18457	90.0	469.31	2.4039E-00	7.7874E-01	4.3688E-09	1.3788E-03	1.1274E-02	7.1728E-04		
42	18876	90.0	475.02	2.1025E-00	7.5149E-01	4.2047E-09	1.3721E-03	1.1409E-02	7.8408E-04		
43	19295	90.0	481.15	2.2821E-00	7.2924E-01	4.4834E-09	1.3799E-03	1.1554E-02	6.7468E-04		
44	19714	90.0	486.41	2.7015E-00	7.8851E-01	4.9133E-09	1.4005E-03	1.1674E-02	7.5252E-04		
45	20133	90.0	498.44	2.4548E-00	7.2488E-01	4.3571E-09	1.0292E-03	1.1779E-02	7.4788E-04		
46	20552	90.0	495.89	4.5555E-00	1.2552E-00	1.5767E-03	6.3070E-03	1.1903E-02	1.2054E-03		
47	20971	90.0	501.54	5.2224E-00	1.4914E-00	1.8668E-03	1.0169E-03	1.2037E-02	1.4429E-03		
48	21390	90.0	504.34	5.4324E-00	1.5242E-00	1.4357E-03	7.8130E-03	1.2108E-02	1.4791E-03		
49	21809	90.0	509.04	5.4727E-00	1.6181E-00	2.0808E-03	3.4600E-03	1.2215E-02	1.5789E-03		
50	22228	90.0	521.84	6.1866E-00	1.5499E-00	2.0125E-03	6.1154E-03	1.2519E-02	1.5364E-03		
51	22647	90.0	525.45	5.7229E-00	1.3477E-00	1.7586E-03	1.0962E-03	1.2409E-02	1.3423E-03		
52	23066	90.0	518.13	4.4150E-00	1.2943E-00	1.6289E-03	6.3379E-03	1.2430E-02	1.2376E-03		
53	23485	90.0	518.14	5.3705E-00	1.3826E-00	1.4880E-03	1.2430E-03	1.2431E-02	1.3036E-03		
54	23904	90.0	519.74	5.1259E-00	1.2839E-00	1.4628E-03	6.7242E-03	1.2488E-02	1.2948E-03		
55	24323	90.0	522.52	5.1922E-00	1.3957E-00	1.4678E-03	6.8426E-03	1.2538E-02	1.2954E-03		
56	24742	90.0	525.84	5.2583E-00	1.2645E-00	1.4844E-03	6.4344E-03	1.2590E-02	1.2582E-03		
57	25161	90.0	529.41	5.0937E-00	1.2249E-00	1.4063E-03	6.4749E-03	1.2498E-02	1.2258E-03		
58	25580	135.0	459.52	7.1093E-01	1.2101E-01	2.5246E-09	1.0215E-03	1.1043E-02	4.9338E-04		
59	26000	135.0	467.54	2.7704E-00	8.0825E-01	1.0774E-03	4.3576E-03	1.1233E-02	6.2496E-04		
60	26419	135.0	472.51	2.7426E-00	6.6666E-01	1.0564E-03	4.2713E-03	1.1349E-02	6.0466E-04		
61	26838	135.0	478.44	2.4530E-00	6.2437E-01	1.0133E-03	6.0945E-03	1.1490E-02	7.7541E-04		
62	27257	135.0	483.44	2.8584E-00	6.4931E-01	1.0999E-03	6.4446E-03	1.1600E-02	6.4143E-04		
63	27676	135.0	498.54	1.3696E-00	9.7904E-01	1.2187E-03	4.9232E-03	1.1729E-02	9.3202E-04		
64	28095	135.0	443.77	4.7908E-00	1.2241E-00	1.5336E-03	6.1930E-03	1.1852E-02	1.1725E-03		
65	28514	135.0	447.85	5.4016E-00	1.5672E-00	1.4735E-03	7.9643E-03	1.1943E-02	1.5085E-03		
66	28933	135.0	502.59	6.4786E-00	1.7469E-00	2.2131E-03	6.9375E-03	1.2001E-02	1.6912E-03		
67	29352	135.0	506.46	6.4368E-00	1.7624E-00	2.2440E-03	6.9569E-03	1.2175E-02	1.7145E-03		
68	29771	135.0	519.56	6.7828E-00	1.6311E-00	2.1118E-03	6.5147E-03	1.2464E-02	1.6125E-03		
69	30190	135.0	522.34	6.1850E-00	1.5034E-00	1.4533E-03	1.7874E-03	1.2429E-02	1.4415E-03		
70	30609	135.0	513.44	1.0944E-00	1.1207E-00	1.4440E-03	5.48110E-03	1.2330E-02	1.1881E-03		
71	31028	135.0	516.22	5.5022E-00	1.4987E-00	2.0809E-03	4.3110E-03	1.2389E-02	1.5738E-03		
72	31447	135.0	518.97	5.2295E-00	1.4609E-00	1.8614E-03	1.5876E-03	1.2424E-02	1.4213E-03		
73	31866	135.0	519.74	5.4302E-00	1.4808E-00	1.4947E-03	1.6248E-03	1.2488E-02	1.4817E-03		
74	32285	135.0	522.52	5.4029E-00	1.3753E-00	1.7874E-03	1.2075E-03	1.2534E-02	1.1645E-03		
75	32704	135.0	510.19	6.0792E-00	1.4619E-00	1.4919E-03	1.7342E-03	1.2717E-02	1.4044E-03		
76	33123	180.0	446.54	1.2858E-00	7.5473E-01	1.1444E-03	4.0442E-03	1.1084E-02	6.7458E-04		
77	33542	180.0	449.55	2.7051E-00	6.1314E-01	1.0088E-03	4.0740E-03	1.1667E-02	7.1044E-04		
78	33961	180.0	479.54	2.7802E-00	6.1594E-01	1.0043E-03	4.0591E-03	1.1515E-02	7.6843E-04		
79	34380	180.0	497.37	4.3400E-00	1.2344E-00	1.5597E-03	6.2940E-03	1.1938E-02	1.1423E-03		
80	34799	180.0	502.59	5.2434E-00	1.3137E-00	1.6641E-03	6.7174E-03	1.2000E-02	1.2717E-03		
81	35218	180.0	507.24	6.2161E-00	1.6507E-00	2.1034E-03	6.4847E-03	1.2171E-02	1.6878E-03		
82	35637	180.0	510.24	5.4626E-00	1.5586E-00	1.4914E-03	6.0340E-03	1.2243E-02	1.4213E-03		
83	36056	180.0	515.49	6.0244E-00	1.5939E-00	2.0541E-03	6.2457E-03	1.2379E-02	1.5888E-03		
84	36475	180.0	522.34	6.9499E-00	1.						

GF ASYMMETRIC TRIM TEST  
HEA TRANSFER PHASE

GROUP 23 SYM PHI XA SYM PHI XA  
MACH NO = 8 0 225  
REF NO = 2.17 X 10\*\*6 45 270  
CONFIG 1 90 315  
ALPHA 1 135  
TIME 0 SEC 180 0.25  
0.25

22-25

24-25

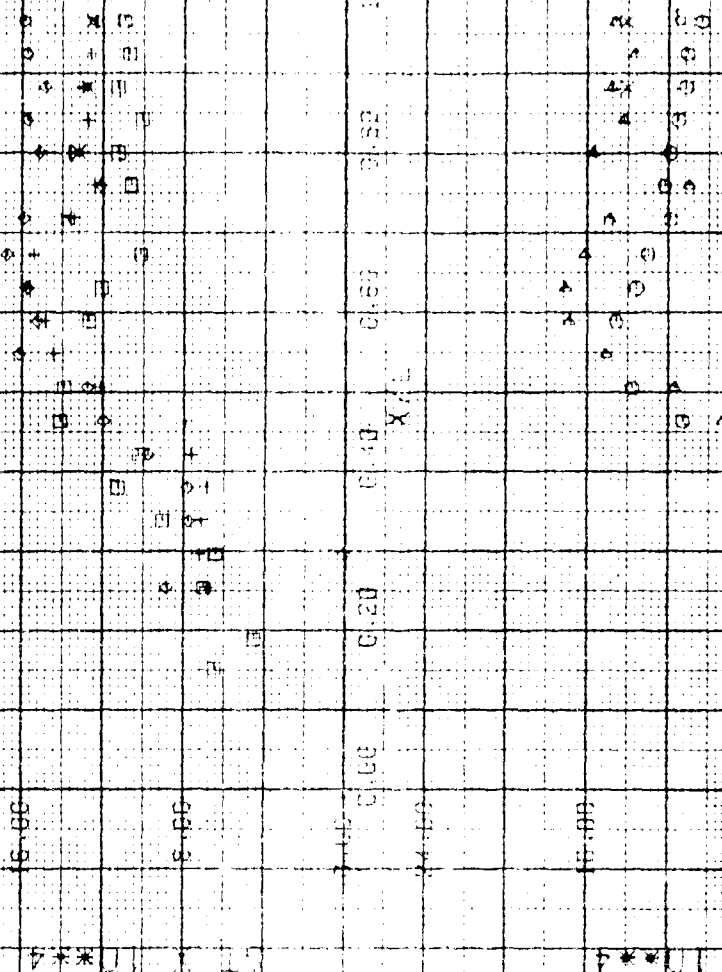
26-25

28-25

30-25

34-25

40-25



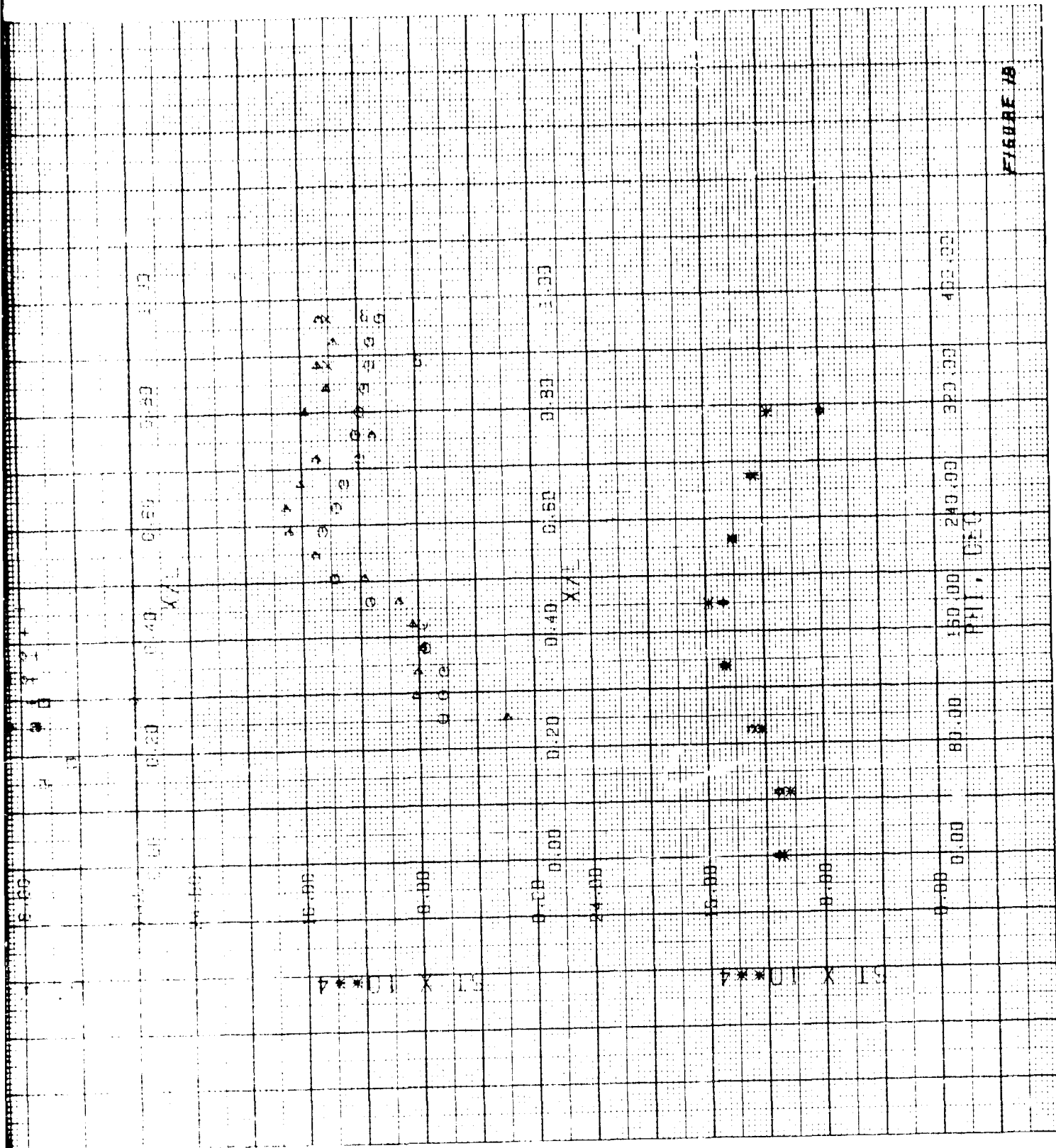


FIGURE 18

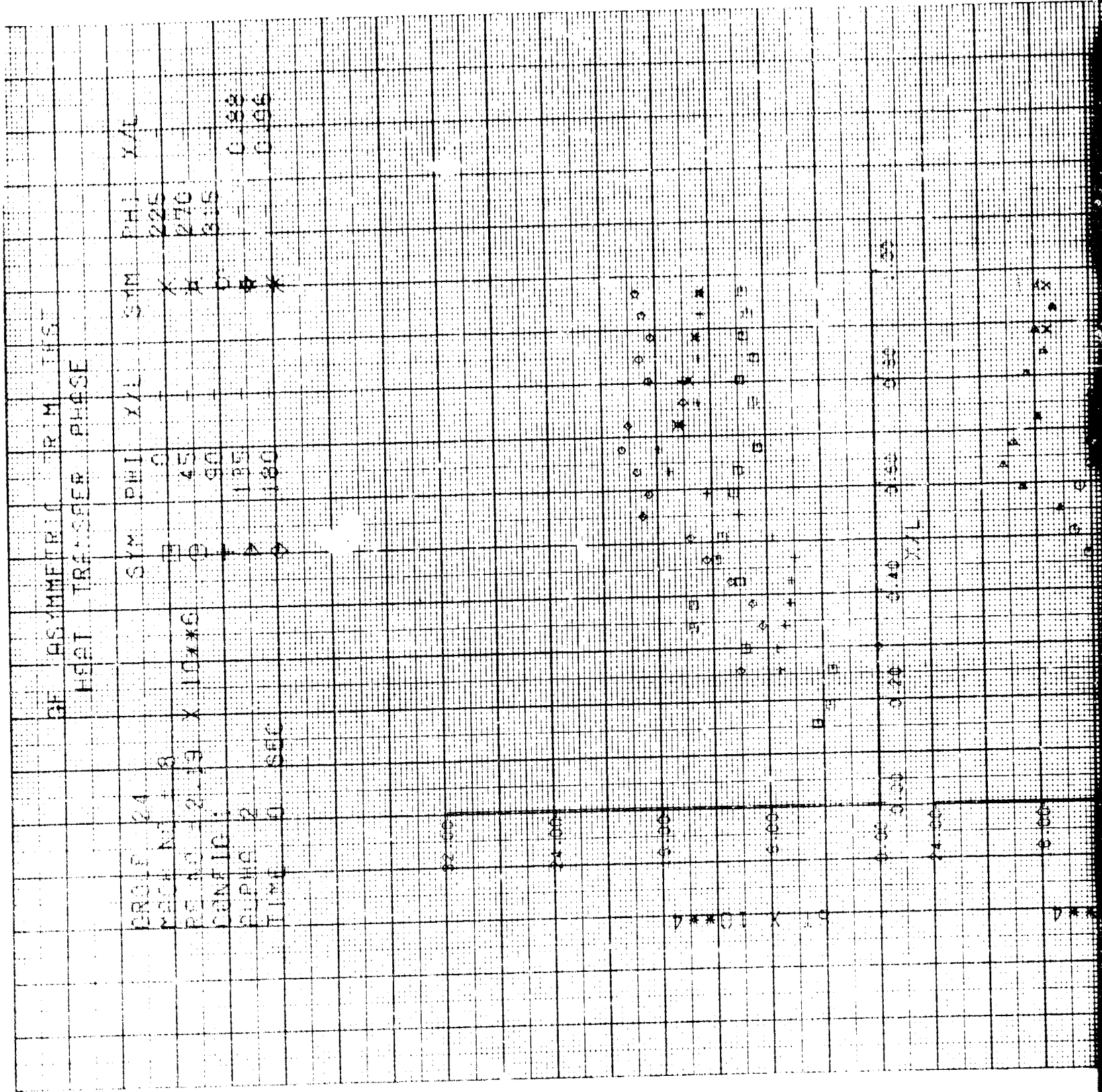
Figure 18

7/16/69

MEDICAL, INC. ANNEX D APT. TENNESSEE  
WIND TUNNEL GAS DYNAMICS FACILITY  
60 INCH HYPERSONIC TUNNEL

Table with columns: GROUP, COMP ID, ALPHA-M, ALPHA-H, ALPHA-L, HULL, ALPHA-R, DATE, MACH NO, TO, TIME. Rows include data points for various test conditions and parameters.

Vertical text on the right side of the page, possibly a stamp or label, partially obscured and difficult to read.



DROLF 24  
 MAP# No 18  
 RE NO 2119 X 10\*\*6  
 CONFIG: 1  
 SLPHS 2  
 TIME 0 SEC

PH1 225  
 PH1 270  
 PH1 315  
 PH1 360  
 PH1 405

SYM X  
 SYM O  
 SYM B  
 SYM A  
 SYM X

PH1 0  
 PH1 45  
 PH1 90  
 PH1 135  
 PH1 180

PH1 X 10\*\*6  
 PH1 X 10\*\*6

2

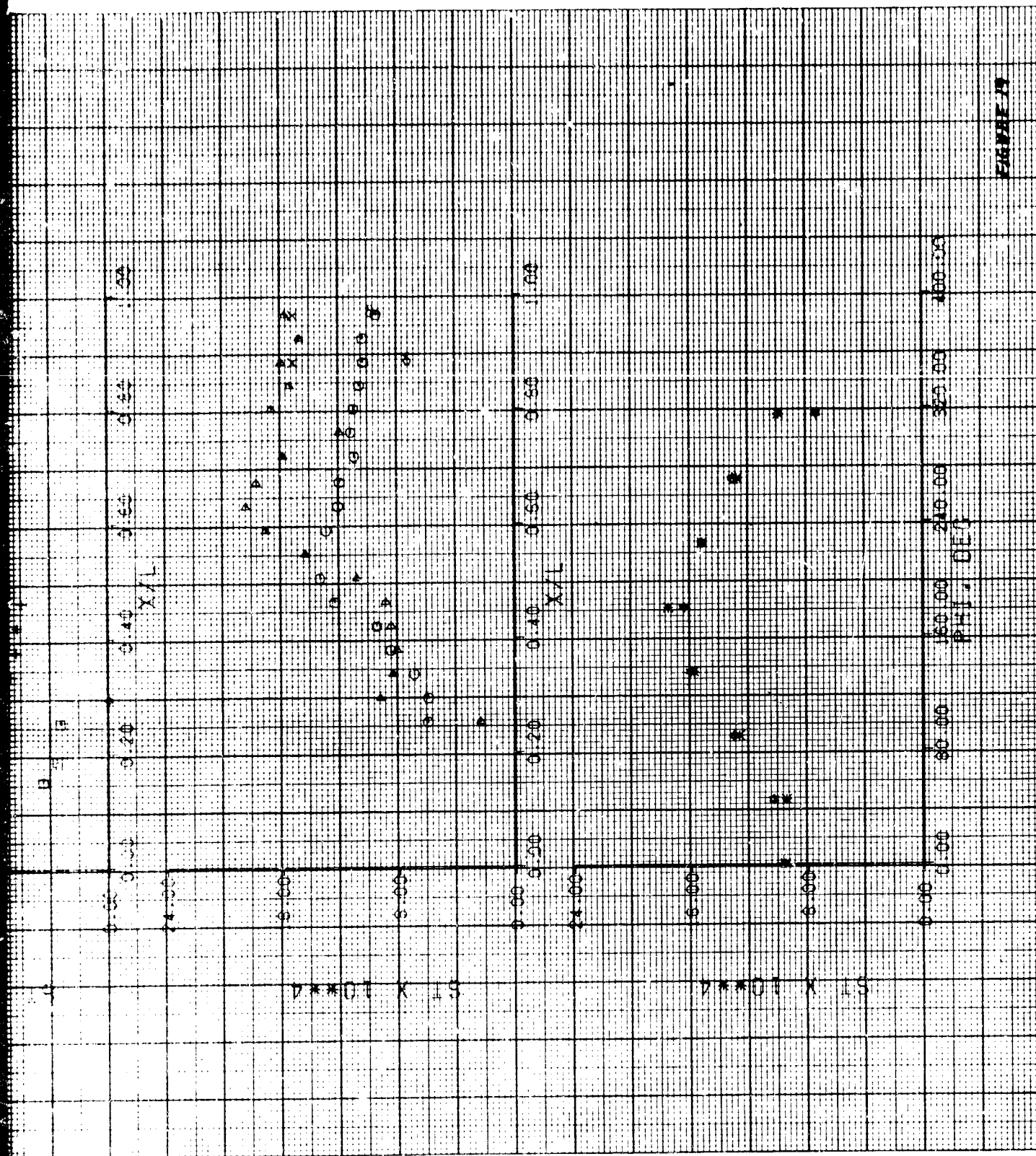


FIGURE 19

Figure 19

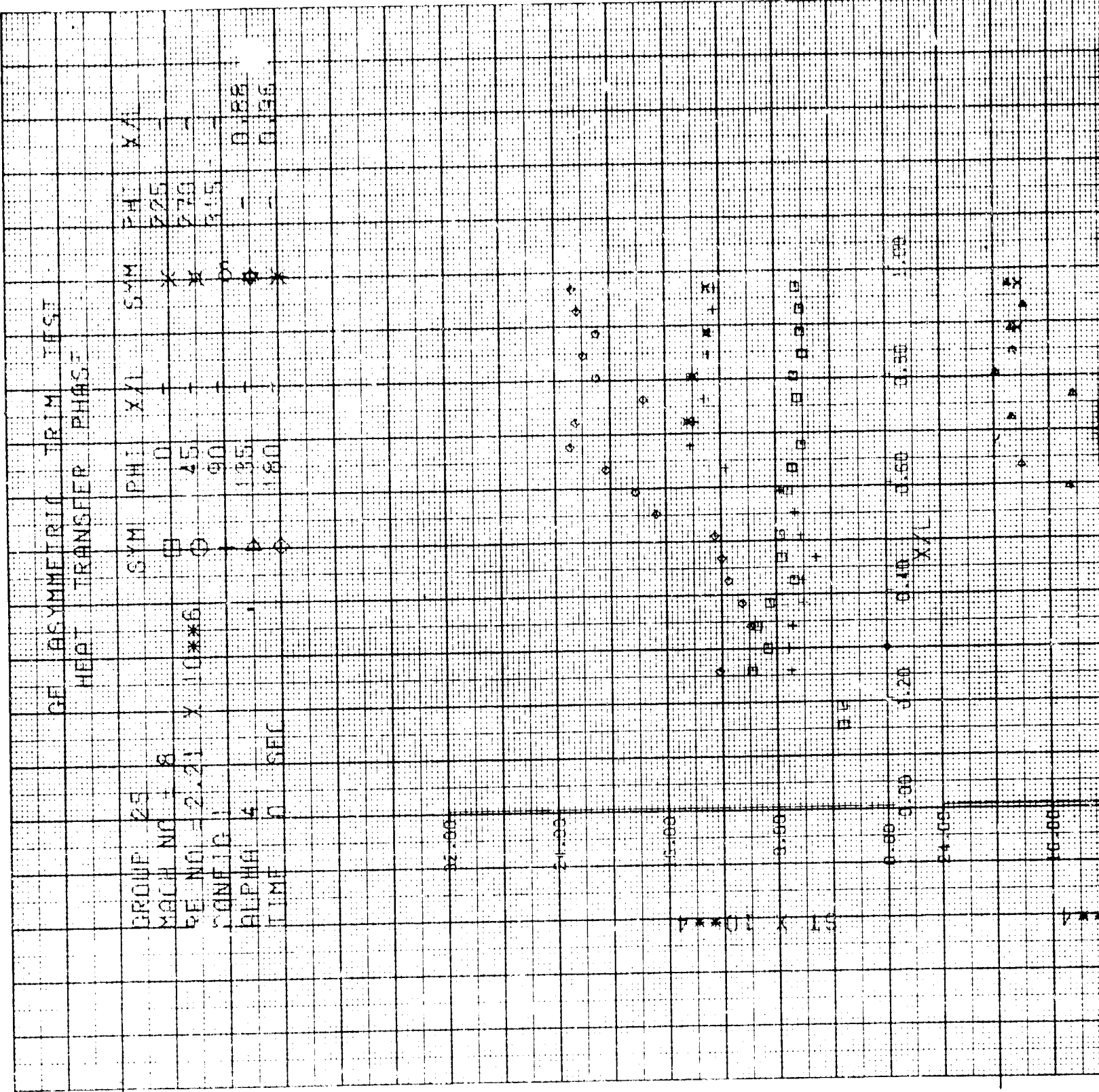
3

7/16/69

AEDCIARD (INC.) ANNULD AFS, TENNESSEE  
VON KARMAN GAS DYNAMICS FACILITY  
50 INCH HYPERSONIC TUNNEL J  
UTOP1-M00

Table with columns: GROUP, COMP ID, ALPHA-H, BETA-H, ALPHA-H, BETA-H, ALPHA-H, BETA-H, DATE, MACH NO, PO, TO, TIME. Rows include numerical data for various parameters across different test cases.

GE ASYMMETRIC TRIM TEST  
HEAT TRANSFER PHASE  
GROUP 25  
MACH NO 1.8



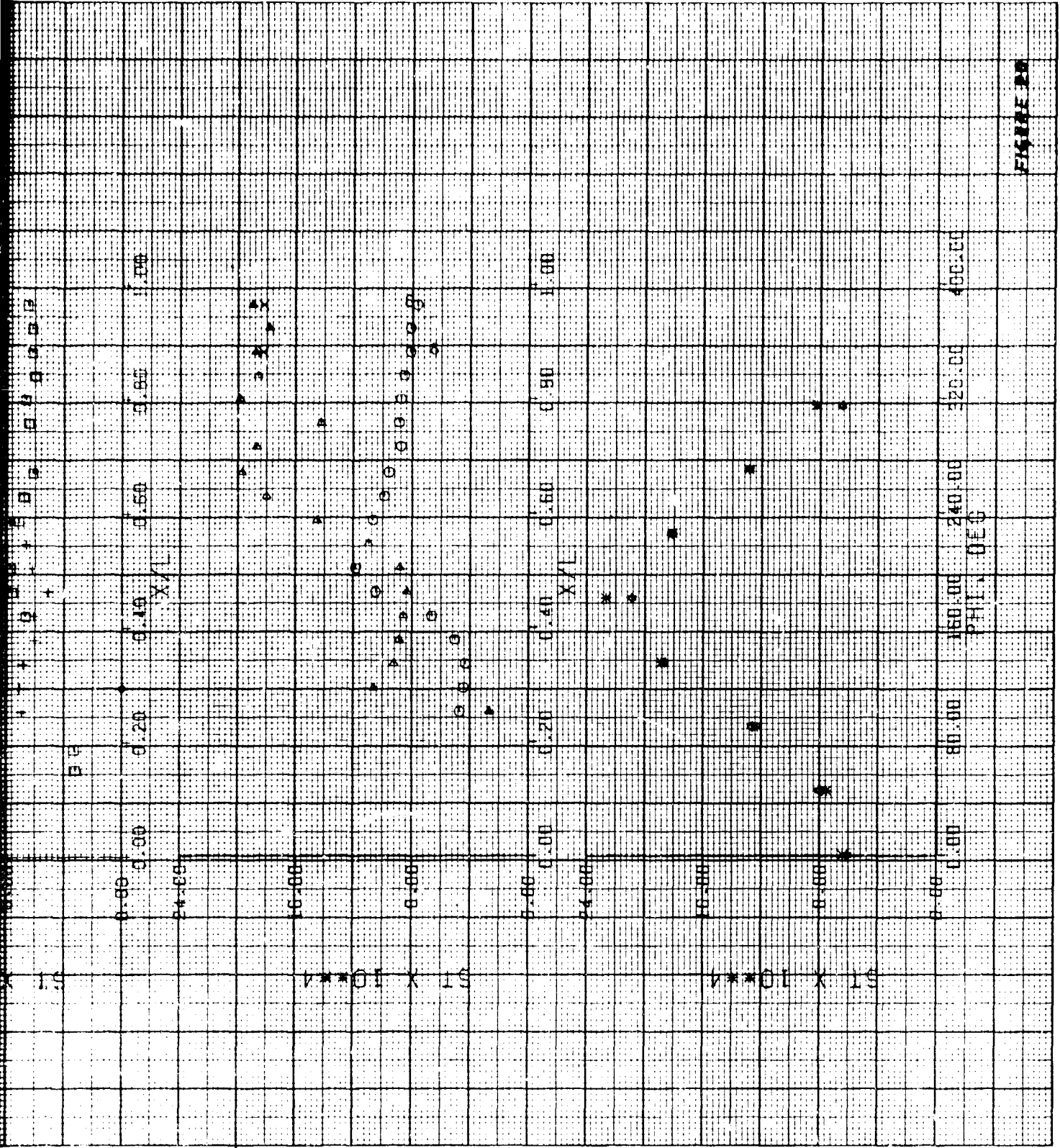


FIGURE 20

Figure 20

3

REDC (ANO. INC.) AMMOLD AFB, TENNESSEE  
VON KARMAN GAS DYNAMICS FACILITY  
30 INCH HYPERSONIC TUNNEL W  
VIOTAF-000

GROUP NO	COMP ID	ALPHA-W	BETA-W	ALPHA-X	MULL	ALPH	DATE	RACH NU	PO	TO	TIME
1	1	.65	0	.65	0	0	714	7.09	468.07	1203.07	-60
PHI	TIMP	SLOPE	0-STON	HT	MM						
RE NO	L/L										
1	1700	0	497.44	2.7423E	00	4.1183E-01	1.0843E-03	0.3799E-03	1.1700E-02	8.2014E-04	
1	1807	0	489.44	2.7423E	00	4.1183E-01	1.0843E-03	0.3799E-03	1.1700E-02	8.2014E-04	
1	2524	0	487.92	2.6036E	00	7.0905E-01	9.6731E-03	2.9300E-03	1.1700E-02	8.2014E-04	
4	2543	0	494.74	2.7894E	00	7.2200E-01	9.1663E-03	1.7035E-03	1.1750E-02	6.9050E-04	
6	3363	0	497.52	2.1961E	00	7.5635E-01	9.9293E-03	3.0000E-03	1.1941E-02	7.3360E-04	
6	3743	0	498.96	2.2760E	00	7.7372E-01	9.8761E-03	3.4870E-03	1.1975E-02	7.5602E-04	
7	4722	0	498.56	2.2181E	00	7.3167E-01	9.3245E-03	3.1765E-03	1.1966E-02	7.1062E-04	
8	4623	0	501.27	1.7570E	00	1.0330E-01	1.1223E-03	2.3940E-03	1.2030E-02	1.0074E-03	
9	5043	0	502.84	4.1303E	00	1.3040E-01	1.6777E-03	0.7745E-03	1.2000E-02	1.2786E-03	
10	5043	0	513.31	8.4030E	02	2.1800E-02	3.1700E-03	1.2401E-03	1.2741E-02	2.9145E-03	
11	5002	0	509.34	5.8200E	00	1.6030E-01	2.0720E-03	9.3002E-03	1.2222E-02	1.5704E-03	
12	6302	0	515.23	3.5300E	00	1.4900E-01	1.4490E-03	7.6000E-03	1.2330E-02	1.6430E-03	
13	6722	0	519.91	6.0542E	00	1.2411E-01	1.6263E-03	6.5070E-03	1.2372E-02	1.2377E-03	
14	6822	315.0	531.24	3.1500E	00	4.2057E-01	1.0715E-03	9.4012E-03	1.2742E-02	0.3015E-04	
15	7503	0	525.41	4.1594E	00	1.2407E-01	1.6376E-03	6.6050E-03	1.2603E-02	1.2550E-03	
16	7900	0	525.37	4.4689E	00	1.1100E-01	1.7200E-03	0.9730E-03	1.2602E-02	1.3252E-03	
17	8449	0	525.17	4.4432E	00	1.1600E-01	1.5317E-03	6.1742E-03	1.2407E-02	1.1456E-03	
18	8922	0	524.77	4.4200E	00	1.2320E-01	1.6340E-03	6.5000E-03	1.2400E-02	1.2430E-03	
19	9235	0	525.94	4.6032E	00	1.2283E-01	1.6310E-03	6.5773E-03	1.2711E-02	1.2400E-03	
20	9648	0	529.56	4.1253E	00	1.2800E-01	1.6900E-03	6.4500E-03	1.2702E-02	1.2427E-03	
21	2524	45.0	445.90	2.5688E	00	7.5937E-01	9.5284E-03	3.0511E-03	1.1400E-02	7.2630E-04	
22	2543	45.0	492.95	2.7232E	00	7.1279E-01	9.0213E-03	3.6453E-03	1.1433E-02	6.8757E-04	
23	3363	45.0	495.63	2.1092E	00	6.9416E-01	8.8153E-03	3.5000E-03	1.1446E-02	6.7170E-04	
24	3743	45.0	497.80	2.2495E	00	7.5103E-01	9.5513E-03	3.0001E-03	1.1930E-02	7.2410E-04	
25	4722	45.0	497.81	2.6344E	00	7.6545E-01	9.7374E-03	3.9340E-03	1.1929E-02	7.4262E-04	
26	4623	45.0	499.91	1.5240E	00	1.0100E-01	1.2400E-03	5.2440E-03	1.1990E-02	9.4020E-04	
27	5043	45.0	502.45	5.2412E	00	1.3910E-01	1.7824E-03	7.2007E-03	1.2050E-02	1.3502E-03	
100	9648	315.0	534.37	4.9165E	00	1.1823E-01	1.5702E-03	6.3007E-03	1.2116E-02	1.2000E-03	
29	5002	45.0	506.81	6.0300E	00	1.6615E-01	2.1404E-03	8.6073E-03	1.2162E-02	1.6301E-03	
30	5302	45.0	512.34	5.8760E	00	1.5350E-01	1.9250E-03	6.4077E-03	1.2250E-02	1.5170E-03	
31	6722	45.0	521.37	5.9830E	00	1.4680E-01	1.9284E-03	7.7801E-03	1.2507E-02	1.4675E-03	
32	7170	45.0	524.72	4.5000E	00	1.3160E-01	1.7353E-03	7.0000E-03	1.2500E-02	1.3200E-03	
33	7503	45.0	523.18	4.4557E	00	1.3379E-01	1.7666E-03	7.1027E-03	1.2550E-02	1.3397E-03	
34	7900	45.0	527.71	4.7160E	00	1.3110E-01	1.7420E-03	6.9547E-03	1.2539E-02	1.3120E-03	
35	8449	45.0	524.87	4.7160E	00	1.2755E-01	1.6623E-03	6.7841E-03	1.2590E-02	1.2900E-03	
36	8922	45.0	527.89	4.4501E	00	1.2255E-01	1.6225E-03	6.5440E-03	1.2600E-02	1.2430E-03	
37	9235	45.0	529.56	4.8055E	00	1.1990E-01	1.5924E-03	6.4231E-03	1.2700E-02	1.2113E-03	
38	9648	45.0	529.57	4.7824E	00	1.1300E-01	1.5000E-03	6.0500E-03	1.2701E-02	1.1416E-03	
39	2524	90.0	489.68	2.1062E	00	6.4263E-01	8.6039E-03	3.0773E-03	1.1755E-02	6.5800E-04	
40	2543	90.0	495.31	2.2036E	00	7.1719E-01	9.1001E-03	3.0743E-03	1.1800E-02	6.9300E-04	
41	3363	90.0	494.43	2.1070E	00	7.2841E-01	9.2590E-03	3.7410E-03	1.1915E-02	7.6562E-04	
42	3743	90.0	495.81	2.4584E	00	7.7595E-01	9.4541E-03	3.9013E-03	1.1940E-02	7.5094E-04	
43	4722	90.0	495.81	2.8760E	00	7.1273E-01	1.1582E-03	6.6766E-03	1.1882E-02	8.2640E-04	
44	4623	90.0	494.48	3.5222E	00	1.0941E-01	1.3675E-03	5.9000E-03	1.1809E-02	1.0574E-03	
45	5043	90.0	495.91	5.2520E	00	1.4471E-01	1.8304E-03	7.4272E-03	1.1993E-02	1.4009E-03	
46	5002	90.0	496.83	5.8112E	00	1.6555E-01	2.1107E-03	8.2070E-03	1.1925E-02	1.6004E-03	
47	5002	90.0	500.47	5.7120E	00	1.6028E-01	2.0481E-03	8.2725E-03	1.2011E-02	1.5000E-03	
48	6302	90.0	506.49	5.7090E	00	1.5951E-01	2.0530E-03	8.2930E-03	1.2154E-02	1.5640E-03	
49	6722	90.0	518.88	5.8965E	00	1.4772E-01	1.9270E-03	7.7802E-03	1.2400E-02	1.4675E-03	
50	7170	90.0	521.47	4.5197E	00	1.2994E-01	1.7067E-03	6.8847E-03	1.2509E-02	1.2900E-03	
51	7503	90.0	515.74	4.7007E	00	1.2245E-01	1.5950E-03	6.4400E-03	1.2373E-02	1.2140E-03	
52	7900	90.0	515.84	4.6906E	00	1.3277E-01	1.7287E-03	6.9770E-03	1.2357E-02	1.3100E-03	
53	8449	90.0	516.81	4.6000E	00	1.2670E-01	1.6520E-03	6.6700E-03	1.2380E-02	1.2500E-03	
54	8922	90.0	519.87	4.1153E	00	1.2557E-01	1.6452E-03	6.6300E-03	1.2471E-02	1.2521E-03	
55	9235	90.0	522.31	5.0776E	00	1.2217E-01	1.6051E-03	6.4754E-03	1.2520E-02	1.2210E-03	
56	9648	90.0	522.37	4.1995E	00	1.2395E-01	1.6270E-03	6.5540E-03	1.2531E-02	1.2500E-03	
57	2524	135.0	486.15	6.1000E	01	1.8050E-01	2.2590E-03	9.1340E-03	1.1630E-02	1.1720E-03	
58	2543	135.0	488.56	4.4950E	00	7.9995E-01	1.0860E-03	6.0000E-03	1.1720E-02	7.6757E-04	
59	3363	135.0	488.17	2.4319E	00	7.9916E-01	1.0954E-03	6.0035E-03	1.1720E-02	7.6645E-04	
60	3743	135.0	486.66	2.4554E	00	7.9310E-01	9.9673E-03	6.0200E-03	1.1684E-02	7.5900E-04	
61	4722	135.0	485.28	3.1497E	00	7.4661E-01	1.1865E-03	6.7940E-03	1.1651E-02	8.6460E-04	
62	4623	135.0	484.45	4.2820E	00	1.1704E-01	1.4659E-03	5.9262E-03	1.1637E-02	1.1170E-03	
63	5043	135.0	485.58	5.7141E	00	1.6600E-01	1.8380E-03	7.4007E-03	1.1650E-02	1.3450E-03	
64	5002	135.0	486.25	6.4195E	00	1.7047E-01	2.1394E-03	8.4000E-03	1.1674E-02	1.4032E-03	
65	5002	135.0	489.29	6.1389E	00	1.6833E-01	2.1260E-03	8.3070E-03	1.1740E-02	1.4040E-03	
66	6302	135.0	495.84	6.1213E	00	1.6255E-01	2.0043E-03	8.3402E-03	1.1947E-02	1.5731E-03	
67	6722	135.0	516.70	5.9873E	00	1.4390E-01	1.8642E-03	7.5245E-03	1.2254E-02	1.4194E-03	
68	7170	135.0	514.21	4.5560E	00	1.3066E-01	1.7020E-03	6.8603E-03	1.2317E-02	1.2957E-03	
69	7503	135.0	505.26	4.3310E	00	1.0180E-01	1.3000E-03	5.2401E-03	1.2125E-02	9.9762E-04	
70	7900	135.0	505.47	4.4016E	00	1.4242E-01	1.8315E-03	7.3450E-03	1.2130E-02	1.3450E-03	
71	8449	135.0	507.57	4.4810E	00	1.3449E-01	1.7342E-03	7.0019E-03	1.2140E-02	1.3207E-03	
72	8922	135.0	511.09	4.6902E	00	1.3614E-01	1.7376E-03	7.0143E-03	1.2260E-02	1.3230E-03	
73	9235	135.0	516.70	5.8457E	00	1.2385E-01	1.6167E-03	6.5143E-03	1.2370E-02	1.2791E-03	
74	9648	135.0	519.87	4.4014E	00	1.2080E-01	1.7092E-03	6.6002E-03	1.2452E-02	1.2460E-03	
75	2524	180.0	485.84	2.4430E	00	7.3911E-01	9.2712E-03	3.7470E-03	1.1600E-02	7.0000E-04	
76	2543	180.0	-116.74	-1.1540E-14	-3.7020E-15	-2.6400E-18	-1.1040E-17	-9.1030E-04	-9.1030E-04	-2.8050E-18	
77	3363	180.0	489.78	2.4200E	00	7.1561E-01	9.0190E-03	3.6000E-03	1.1560E-02	6.8750E-04	
78	3743	180.0	488.13	2.4031E	00	8.0553E-01	1.0146E-03	6.1000E-03	1.1710E-02	7.7400E-04	
79	4722	180.0	487.88	3.1850E	00	1.0305E-01	1.2957E-03	5.2133E-03	1.1711E-02	8.8700E-04	
80	4623	180.0	488.82	4.7840E	00	1.3662E-01	1.7194E-03	6.9440E-03	1.1716E-02	1.3100E-03	
81	5043	180.0	488.83	5.7179E	00	1.4324E-01	1.4071E-03	7.2870E-03	1.1731E-02	1.3740E-03	
82	5002	180.0	489.17	6.4230E	00	1.7050E-01	2.1406E-03	8.0030E-03	1.1743E-02	1.4070E-03	
83	5002	180.0	491.46	5.4512E	00	1.4741E-01	1.8620E-03	7.5240E-03	1.1760E-02	1.4190E-03	
84	6302	180.0	499.97	5.3450E	00	1.4194E-01	1.8125E-03	7.3212E-03	1.1999E-02	1.3800E-03	
85	6722	180.0	515.99	5.0771E	00	1.4374E-01	1.8737E-03	7.5610E-03	1.2374E-02	1.4261E-03	
86	7170	180.0	519.87	4.6934E							

GE ASYMMETRIC PRIM TEST

HEGT TRANSFER PHASE

GROUP 26  
 MACH NO 8  
 RH NO 2.20 X 10\*\*6  
 CONFIG 1  
 BURH 0  
 TIME C SEC

SYM PH1 XL  
 X 225  
 H 210  
 D 310  
 P 0  
 X 0

SYM PH1 XL  
 C 45  
 D 50  
 P 135  
 X 180

02:00

04:00

06:00

08:00

09:00

09:00

09:00

10\*\*6 X 10\*\*6

0.2

0.4

0.6

0.8

0.55

0.50

0.40

0.30

X/E

0.2

0.4

0.6

0.8

0.9

0.95

0.98

1.0

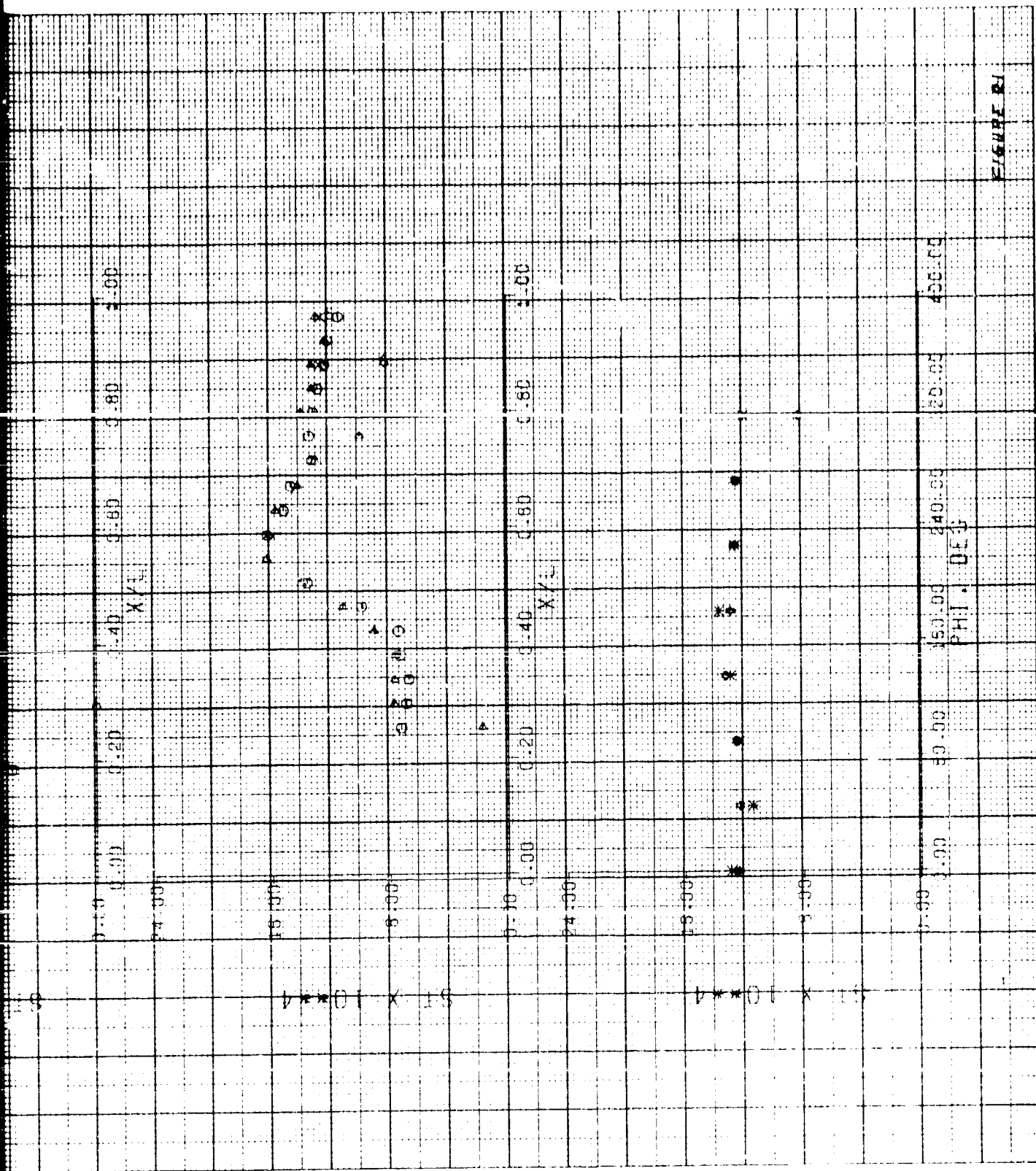


FIGURE 21

Figure 21

7/16/68

MEDICAL INC. DIVISION OF S. TENNESSEE  
VAN HANMAN GAS DYNAMICS FACILITY  
99 INCH HYPERSONIC TUNNEL  
VIDEOT-000

Table with columns: GROUP, COMP ID, ALPHA-M, MEAS-M, ALPHA-S, MULL, ALPH-D, DATE, MACH NO, P, TO, TIME. Rows contain numerical data for various test parameters.

Large grid table with columns: X, Y, Z, etc. Contains numerical data for spatial coordinates and other variables.

SYMMETRIC TRIM TEST  
HEAT TRANSFER PLEASE  
SYMM PHIL Y/L SYM PHL X/L SYM PHL X/L

GROUP 52

GE ASYMMETRIC IRIM TEST  
HEAT TRANSFER PHASE

CRUCIF 52  
 MACH NO 8  
 RE NO 2.15 X 10\*\*6  
 CONFIG 1  
 ALPHA 0  
 TIME 0 SEC

SYM PHI X/L SYM PHI X/L  
 \* 225 \*  
 \* 270 \*  
 \* 315 \*  
 \* 0.88 \*  
 \* 0.96 \*

24-00

27-00

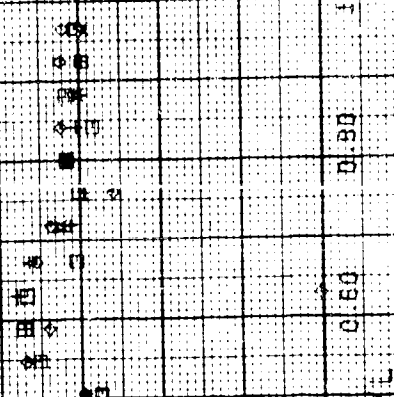
30-00

33-00

36-00

39-00

42-00



ST X 10\*\*4

ST X 10\*\*4

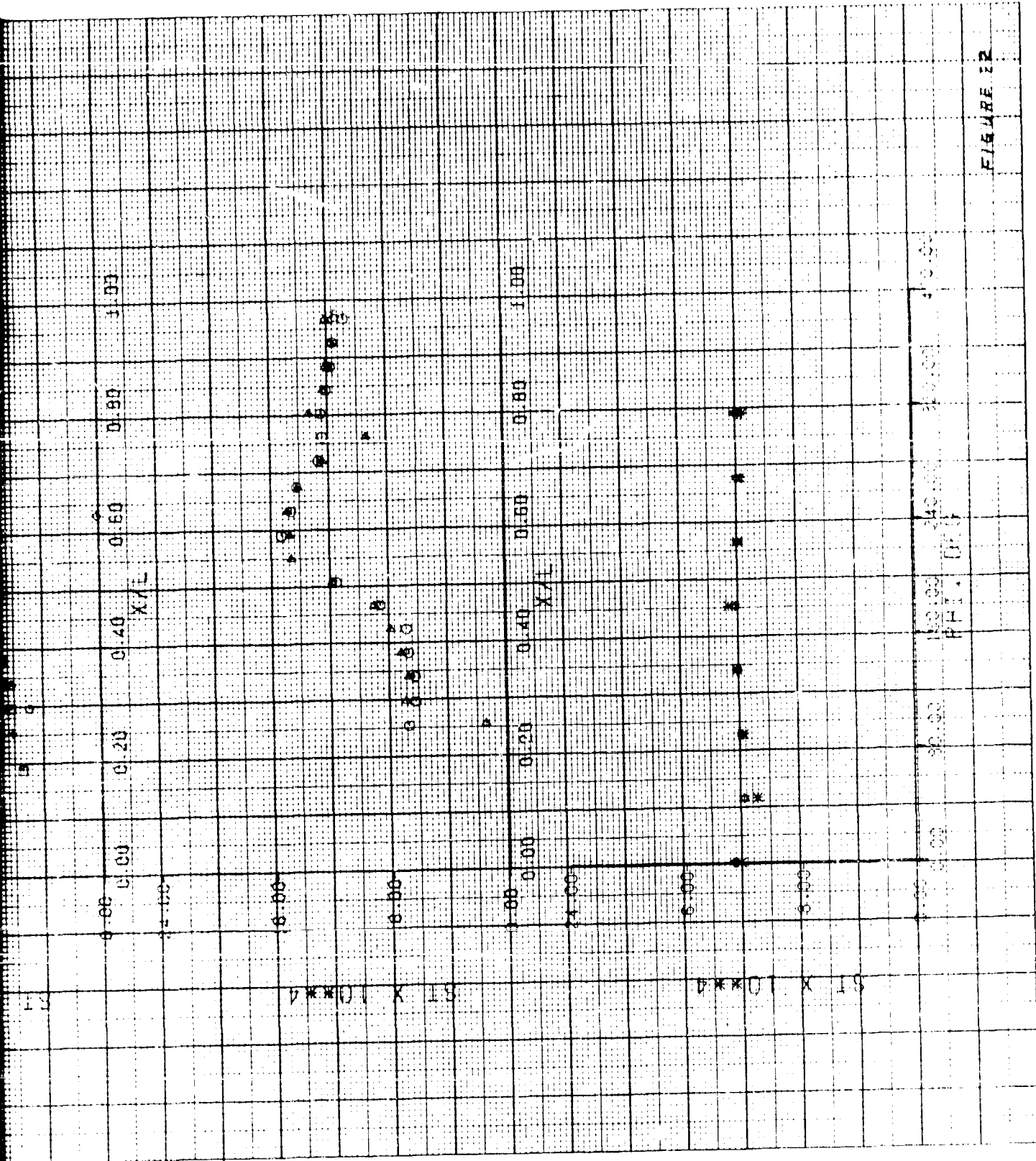


FIGURE 22

Figure 22

AEDC (ARO-INC.) ARNOLD AFB, TENNESSEE  
VON KARMAN GAS DYNAMICS FACILITY  
50 INCH HYPERSONIC TUNNEL B

GROUP	CONF ID	ALPHA-H	BETA-H	ALPHA-S	ROLL	ALPHA-P	DATE	MACH NO	PO	TM	TIME
T-IMP	P-IMP	B-IMP	V-IMP	RHO-IMP	HU-IMP	DU-IMP	RE/FI	RE/FI	RE/D	RE/D	RE/D
9.38400E-01	2.94701E-02	1.31030E-00	3.78310E-03	2.63400E-05	7.55500E-06	1.11907E-06	1.21029E-06	3.14688E-02			
TC NO	X/C	PHI	TEMP	SLOPE	W-STON	HT	HM	HM	ST		
1	1156	0	515.14	2.4510E-01	3.7607E-01	7.0020E-04	3.0969E-03	1.2835E-02	9.6564E-04		
2	1867	0	515.07	1.8030E-01	3.9595E-01	9.2720E-04	2.1254E-03	1.2835E-02	6.6273E-04		
3	2924	0	514.94	2.4111E-01	7.1333E-01	9.4536E-04	3.8289E-03	1.2822E-02	1.1933E-03		
4	2963	0	513.94	1.1897E-01	4.1054E-01	9.8122E-04	2.1509E-03	1.2806E-02	7.1305E-04		
5	3363	0	513.04	1.3757E-01	4.3022E-01	9.7716E-04	2.3267E-03	1.2799E-02	7.1305E-04		
6	3763	0	513.07	1.4534E-01	4.3009E-01	9.8008E-04	2.3916E-03	1.2805E-02	7.3014E-04		
7	4203	0	512.01	1.4036E-01	3.7002E-01	9.6349E-04	2.0297E-03	1.2790E-02	6.3287E-04		
8	4623	0	514.49	1.4010E-01	3.6623E-01	9.1394E-04	2.0717E-03	1.2819E-02	6.4597E-04		
9	4643	0	514.84	1.1910E-01	3.7633E-01	9.8100E-04	2.0199E-03	1.2827E-02	6.2970E-04		
10	5463	0	516.73	1.4473E-01	4.0610E-01	9.4199E-04	2.1845E-03	1.2872E-02	6.8115E-04		
11	5862	0	518.24	1.4700E-01	4.3257E-01	9.7852E-04	2.3315E-03	1.2908E-02	7.2700E-04		
12	6302	0	548.17	1.6148E-01	4.3742E-01	9.8649E-04	2.3634E-03	1.2954E-02	7.3693E-04		
13	6722	0	541.34	1.7962E-01	4.4944E-01	4.0427E-04	2.4349E-03	1.2992E-02	7.5423E-04		
14	8822	315.0	513.05	1.3553E-01	4.7406E-01	1.1925E-03	4.8023E-03	1.3261E-02	1.4974E-03		
15	7943	0	545.58	2.7022E-01	4.4901E-01	0.7765E-04	3.5357E-03	1.3003E-02	1.1025E-03		
16	7996	0	547.19	3.0259E-01	4.4901E-01	1.1492E-03	4.6290E-03	1.3121E-02	1.4434E-03		
17	8409	0	547.79	3.1310E-01	4.0000E-01	1.0837E-03	4.3652E-03	1.3136E-02	1.3611E-03		
18	8822	0	548.70	3.6823E-01	9.2249E-01	1.2512E-03	5.0395E-03	1.3157E-02	1.9714E-03		
19	9235	0	548.54	3.4099E-01	8.7924E-01	1.1924E-03	4.8027E-03	1.3154E-02	1.4975E-03		
20	9648	0	547.20	3.7372E-01	8.9871E-01	1.2145E-03	4.9802E-03	1.3122E-02	1.5279E-03		
21	2524	45.0	513.06	1.4971E-01	4.7212E-01	6.2703E-04	2.5278E-03	1.2745E-02	7.8818E-04		
22	2943	45.0	512.89	1.4403E-01	4.6179E-01	6.1318E-04	2.4720E-03	1.2781E-02	7.7978E-04		
23	3363	45.0	513.01	1.4611E-01	4.3921E-01	5.8331E-04	2.3513E-03	1.2784E-02	7.3322E-04		
24	3763	45.0	513.41	1.4422E-01	4.3357E-01	5.7604E-04	2.3222E-03	1.2793E-02	7.2468E-04		
25	4203	45.0	514.53	1.7008E-01	3.4891E-01	4.6430E-04	1.8716E-03	1.2820E-02	8.9358E-04		
26	4623	45.0	516.74	1.8011E-01	3.8722E-01	5.1081E-04	2.0830E-03	1.2873E-02	6.3495E-04		
27	5043	45.0	517.63	1.8234E-01	3.7795E-01	5.0595E-04	2.0270E-03	1.2850E-02	6.3495E-04		
28	5463	45.0	551.51	3.4401E-01	8.2943E-01	1.1293E-03	4.8479E-03	1.3224E-02	1.4181E-03		
29	5882	315.0	540.44	1.8772E-01	4.3458E-01	9.8290E-04	2.3489E-03	1.2961E-02	7.3242E-04		
30	6302	45.0	542.44	1.7355E-01	4.7061E-01	6.3291E-04	2.5502E-03	1.3008E-02	7.9517E-04		
31	6722	45.0	544.74	2.1552E-01	5.2903E-01	7.1349E-04	2.8753E-03	1.3063E-02	8.9655E-04		
32	7170	45.0	546.72	2.4517E-01	6.0890E-01	8.1281E-04	3.2743E-03	1.3110E-02	1.0210E-03		
33	7593	45.0	548.05	2.4762E-01	7.3982E-01	9.6189E-04	3.8745E-03	1.3142E-02	1.2081E-03		
34	7996	45.0	549.36	3.0281E-01	8.4964E-01	1.1534E-03	4.6450E-03	1.3173E-02	1.4485E-03		
35	8409	45.0	550.72	3.1824E-01	8.6977E-01	1.1709E-03	4.7159E-03	1.3205E-02	1.4705E-03		
36	8822	45.0	551.55	3.3225E-01	6.7390E-01	1.1899E-03	4.7919E-03	1.3225E-02	1.4942E-03		
37	9235	45.0	551.80	3.4093E-01	6.7111E-01	1.1866E-03	4.7878E-03	1.3233E-02	1.4900E-03		
38	9648	45.0	551.14	3.2944E-01	7.9224E-01	1.0781E-03	4.3419E-03	1.3216E-02	1.3538E-03		
39	2524	90.0	516.50	1.4765E-01	4.3047E-01	5.8240E-04	2.3474E-03	1.2868E-02	7.3194E-04		
40	2943	90.0	516.31	1.4080E-01	4.0349E-01	6.4494E-04	2.5995E-03	1.2863E-02	6.1055E-04		
41	3363	90.0	515.99	1.3540E-01	4.2738E-01	5.6930E-04	2.2968E-03	1.2855E-02	7.1617E-04		
42	3763	90.0	516.41	1.3257E-01	4.1042E-01	5.5835E-04	2.2505E-03	1.2869E-02	7.0172E-04		
43	4203	90.0	517.34	1.4235E-01	4.2081E-01	5.6212E-04	2.2650E-03	1.2888E-02	7.0642E-04		
44	4623	90.0	518.51	1.2643E-01	3.9274E-01	5.2481E-04	2.1120E-03	1.2867E-02	6.5856E-04		
45	5043	90.0	518.12	1.4266E-01	4.2045E-01	5.8220E-04	2.2659E-03	1.2905E-02	7.0649E-04		
46	5463	90.0	540.38	1.4234E-01	4.3957E-01	5.8117E-04	2.3511E-03	1.2959E-02	7.3492E-04		
47	5882	90.0	541.22	1.6436E-01	4.8116E-01	6.1916E-04	2.4951E-03	1.2979E-02	7.7790E-04		
48	6302	90.0	542.92	1.8907E-01	5.9050E-01	7.9125E-04	3.1888E-03	1.2944E-02	9.9425E-04		
49	6722	90.0	543.57	2.1649E-01	5.4284E-01	6.3215E-04	2.9497E-03	1.3059E-02	1.1975E-04		
50	7170	90.0	546.75	2.4531E-01	6.2479E-01	8.4516E-04	3.4464E-03	1.3111E-02	1.0616E-03		
51	7593	90.0	548.28	2.7741E-01	7.2317E-01	9.7763E-04	3.9383E-03	1.3160E-02	1.2288E-03		
52	7996	90.0	547.50	3.2583E-01	8.4671E-01	1.1465E-03	4.6184E-03	1.3129E-02	1.4481E-03		
53	8409	90.0	547.90	3.1054E-01	8.4811E-01	1.1449E-03	4.6289E-03	1.3140E-02	1.4433E-03		
54	8822	90.0	549.11	3.5571E-01	8.7317E-01	1.1649E-03	4.7272E-03	1.3167E-02	1.4882E-03		
55	9235	90.0	548.31	3.4231E-01	8.7129E-01	1.1811E-03	4.7575E-03	1.3148E-02	1.4834E-03		
56	9648	90.0	546.19	3.9940E-01	8.6408E-01	1.1683E-03	4.7062E-03	1.3098E-02	1.4675E-03		
57	2524	135.0	511.69	2.9116E-01	4.5661E-01	1.1489E-04	4.6319E-04	1.2752E-02	1.4443E-04		
58	2943	135.0	511.77	1.9410E-01	4.7419E-01	6.5523E-04	2.6415E-03	1.2754E-02	8.2369E-04		
59	3363	135.0	510.59	1.4227E-01	4.8062E-01	6.3624E-04	2.5652E-03	1.2726E-02	7.9986E-04		
60	3763	135.0	528.06	1.3561E-01	1.2127E-01	5.5580E-04	2.2412E-03	1.2666E-02	6.9884E-04		
61	4203	135.0	530.33	1.4462E-01	1.3472E-01	5.7527E-04	2.3194E-03	1.2720E-02	7.2323E-04		
62	4623	135.0	530.88	1.4119E-01	4.1025E-01	5.4329E-04	2.1904E-03	1.2733E-02	6.8360E-04		
63	5043	135.0	531.67	1.4496E-01	3.9594E-01	5.2488E-04	2.1161E-03	1.2752E-02	6.5983E-04		
64	5463	135.0	532.66	1.6348E-01	4.3411E-01	5.6209E-04	2.3229E-03	1.2774E-02	7.2431E-04		
65	5882	135.0	534.26	1.7088E-01	4.9371E-01	6.0356E-04	2.4330E-03	1.2814E-02	7.5453E-04		
66	6302	135.0	535.96	1.9384E-01	5.1395E-01	6.8522E-04	2.7618E-03	1.2854E-02	8.4119E-04		
67	6722	135.0	539.72	2.4585E-01	5.9050E-01	7.9125E-04	3.1888E-03	1.2944E-02	9.9425E-04		
68	7170	135.0	546.91	2.8575E-01	6.7292E-01	9.6313E-04	3.6393E-03	1.2972E-02	1.1348E-03		
69	7593	135.0	539.31	2.6881E-01	5.8827E-01	7.7713E-04	3.1318E-03	1.2934E-02	9.7653E-04		
70	7996	135.0	540.94	3.0958E-01	6.4892E-01	1.2736E-03	5.1322E-03	1.2973E-02	1.4603E-03		
71	8409	135.0	541.48	3.1493E-01	6.4727E-01	1.1655E-03	4.6963E-03	1.2995E-02	1.4643E-03		
72	8822	135.0	541.86	3.4035E-01	6.4784E-01	1.1273E-03	5.1124E-03	1.2994E-02	1.6003E-03		
73	9235	135.0	542.21	3.4063E-01	6.6069E-01	1.1572E-03	4.6626E-03	1.3003E-02	1.4538E-03		
74	9648	135.0	541.86	3.7851E-01	6.2189E-01	1.2144E-03	4.8945E-03	1.3042E-02	1.5262E-03		
75	2524	180.0	532.50	1.7023E-01	4.9443E-01	6.5644E-04	2.6464E-03	1.2774E-02	8.4119E-04		
76	2943	180.0	533.20	1.0542E-01	1.3799E-01	4.4798E-04	1.8100E-03	1.2788E-02	5.4437E-04		
77	3363	180.0	533.11	1.1606E-01	4.0740E-01	5.3580E-04	2.1660E-03	1.2786E-02	6.7350E-04		
78	3763	180.0	532.95	1.4296E-01	4.2945E-01	5.7028E-04	2.2990E-03	1.2782E-02	7.1685E-04		
79	4203	180.0	533.43	1.5094E-01	4.4621E-01	5.9307E-04	2.3988E-03	1.2798E-02	7.4547E-04		
80	4623	180.0	534.16	1.4913E-01	4.4444E-01	6.0430E-04	2.4365E-03	1.2811E-02	7.4973E-04		
81	5043	180.0	535.29	1.5601E-01	3.9838E-01	5.2661E-04	2.0985E-03	1.2838E-02	6.5433E-04		
82	5463	180.0	536.63	1.6133E-01	4.2842E-01	5.7171E-04	2.3043E-03	1.2872E-02	7.1851E-04		
83	5882	180.0	538.28	1.4623E-01	3.8830E-01	5.1431E-04	2.0929E-03	1.2909E-02	6.5268E-04		
84	6302	180.0	538.80	1.2756E-01	3.3873E-01	4.5217E-04	1.8223E-03	1.2874E-02	5.6820E-04		
85	6722	180.0	545.17	2.4027E-01	5.7781E-01	7.7994E-04	3.1442E-03	1.3			

2

EF ASYMMETRIC TRIM 1651  
HBF TRANSFER PHASE

GROUP B2  
 MODN VFI 1 K  
 RE NF = 11.22 X 10\*\*5  
 CONST = 1  
 SUPHS = 0  
 TIME = 0 SUT

SYM PHI XZ EYM PHI XZ  
 \* \* \* \* \*  
 \* \* \* \* \*  
 \* \* \* \* \*  
 \* \* \* \* \*  
 \* \* \* \* \*

DOWN

0.1  
0.2  
0.3

0.40 0.50  
 XZ  
 0.60  
 0.70  
 0.80  
 0.90  
 1.00

0.00 0.20 0.40 0.60 0.80 1.00  
 XZ

DOWN

0.1  
0.2  
0.3

0.40 0.50  
XZ  
0.60  
0.70  
0.80  
0.90  
1.00

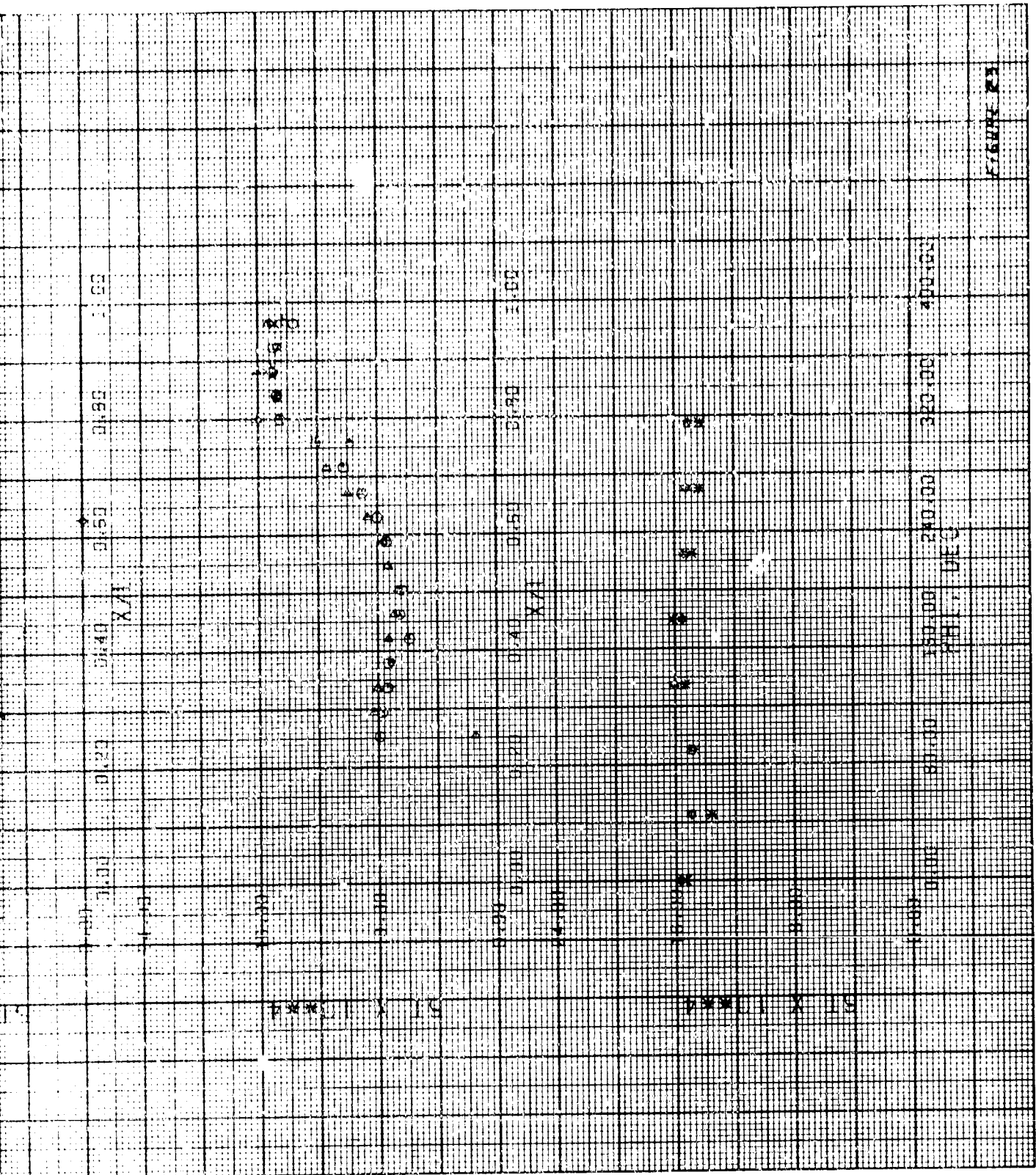
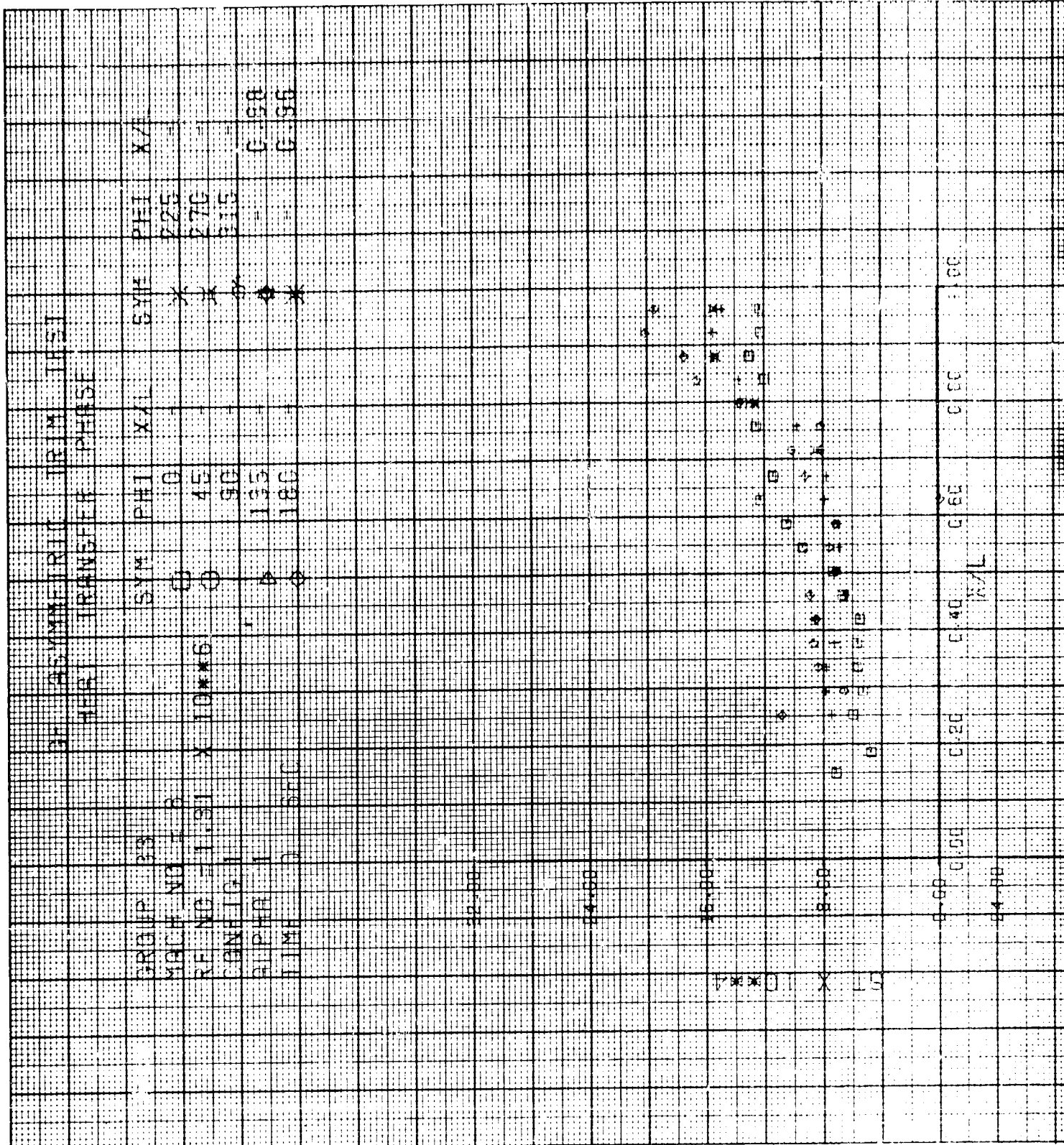


Figure 23

W





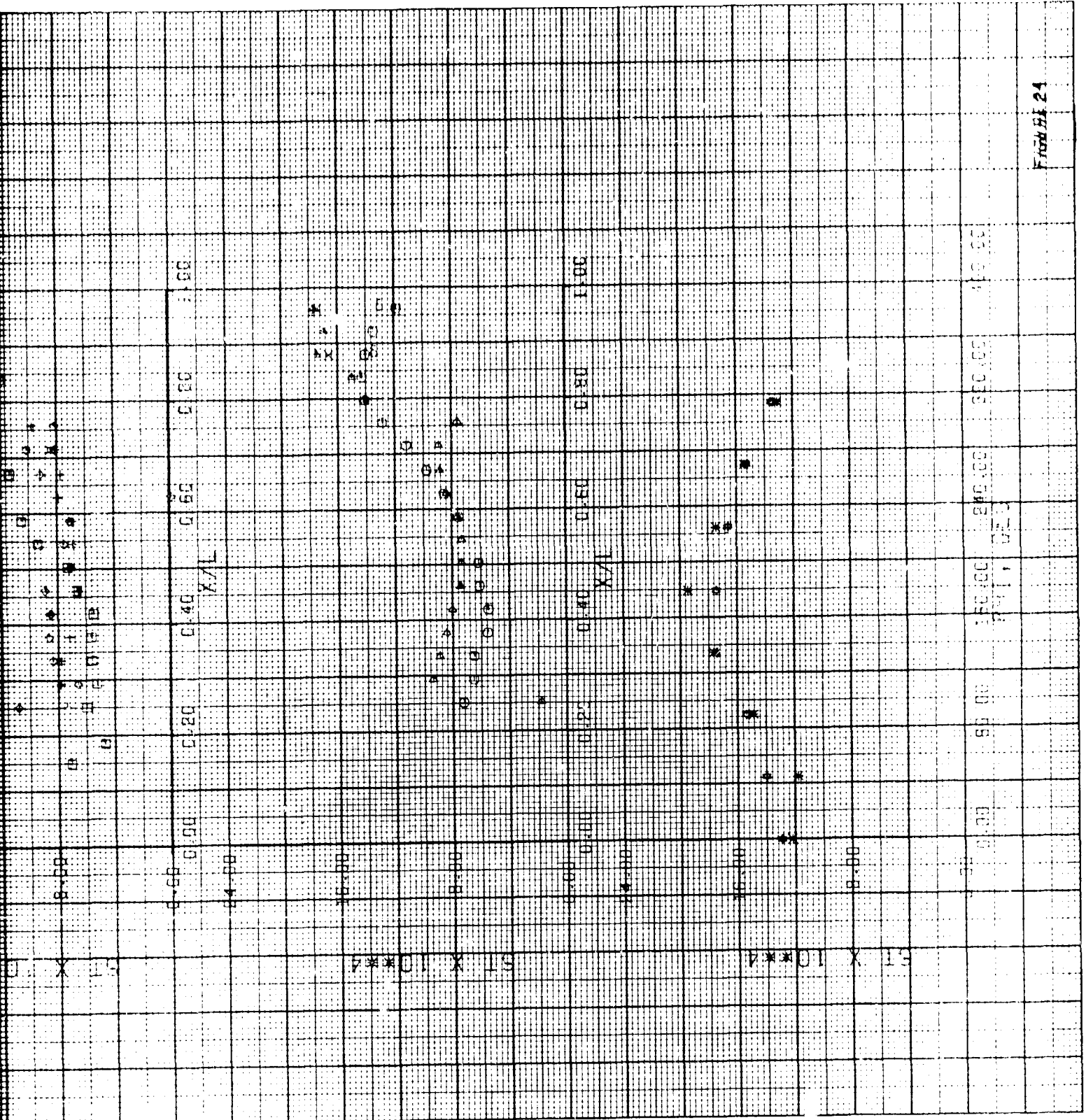


FIGURE 24

Figure 24

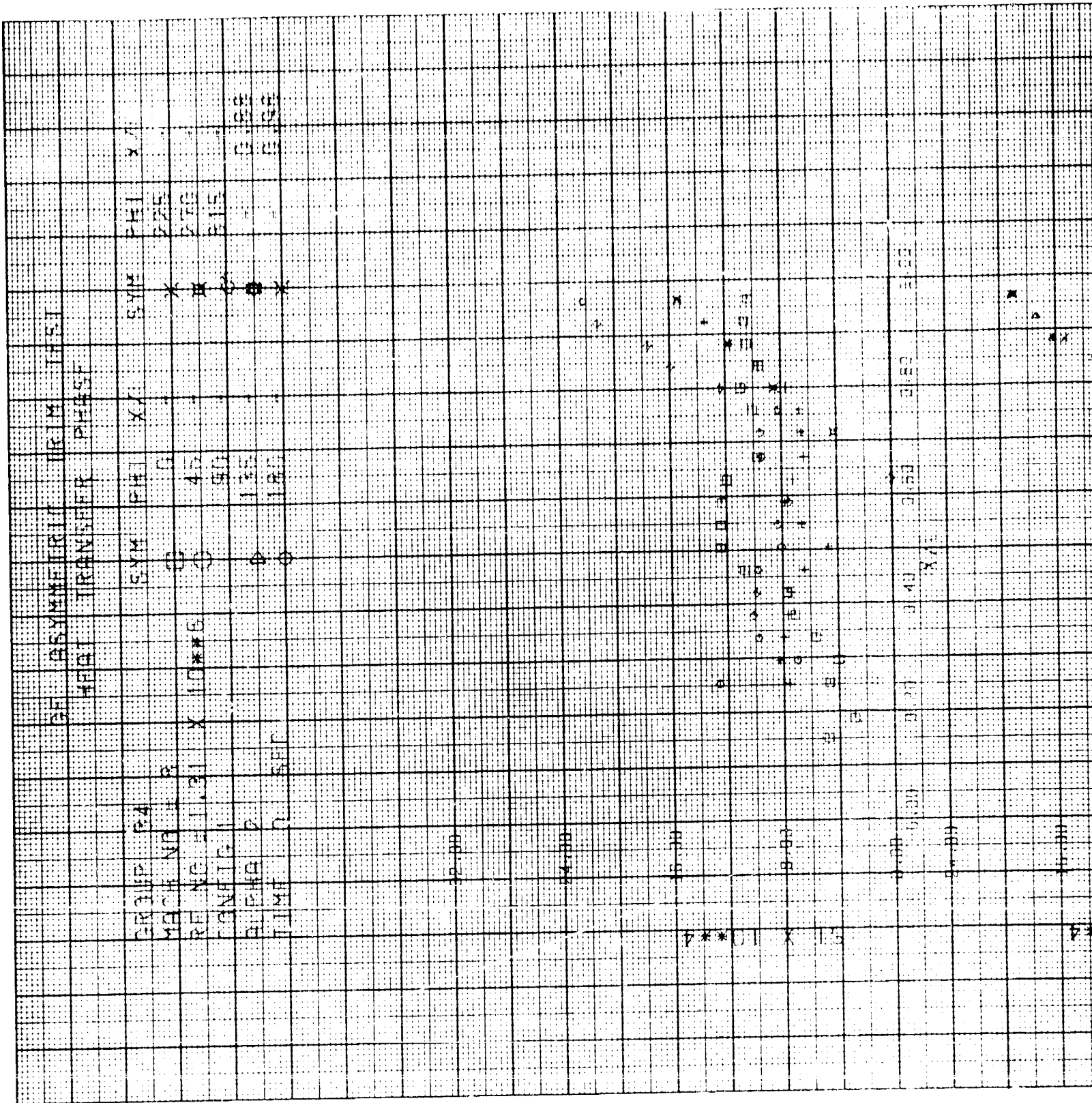
7/14/69

AECIARD, INC. 1 ANNHOLD AVENUE TENNESSEE  
VON KARMAN GAS DYNAMICS FACILITY  
50 INCH HYPERSONIC TUNNEL W  
VT0967-B00

Table with columns: GROUP, COMPID, ALPHA-N, BETA-N, ALPHA-S, RULL, ALP-P, DATE, MACH NO, PD, TO, TIME. Sub-headers include L, P, Q, V, RHO, MU, RE, F, ME, MD, N, S. The table contains 98 rows of numerical data.

Vertical grid on the right side of the page, possibly a secondary data table or a reference table, containing multiple columns and rows of data.

2



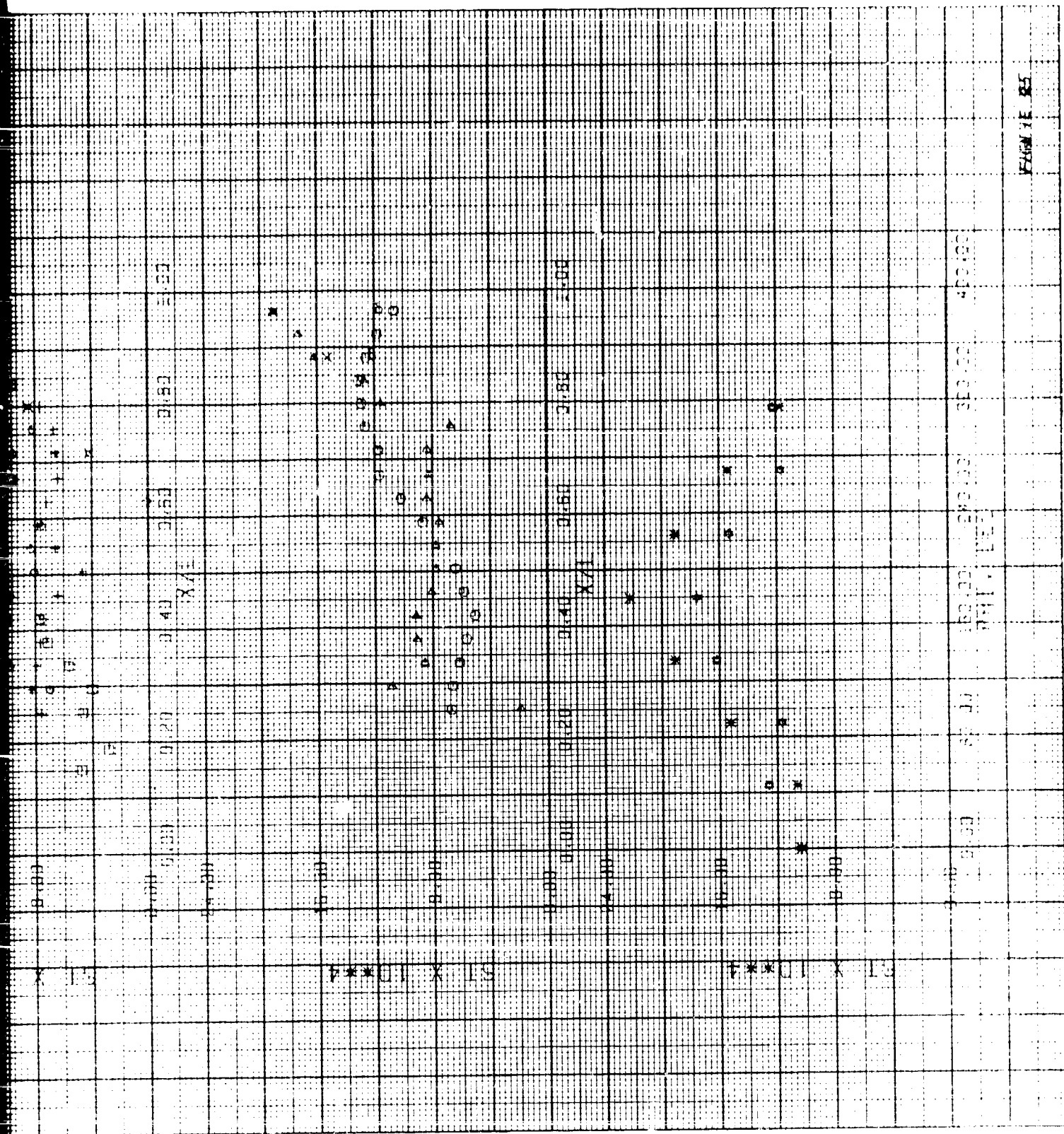


Figure 28

61

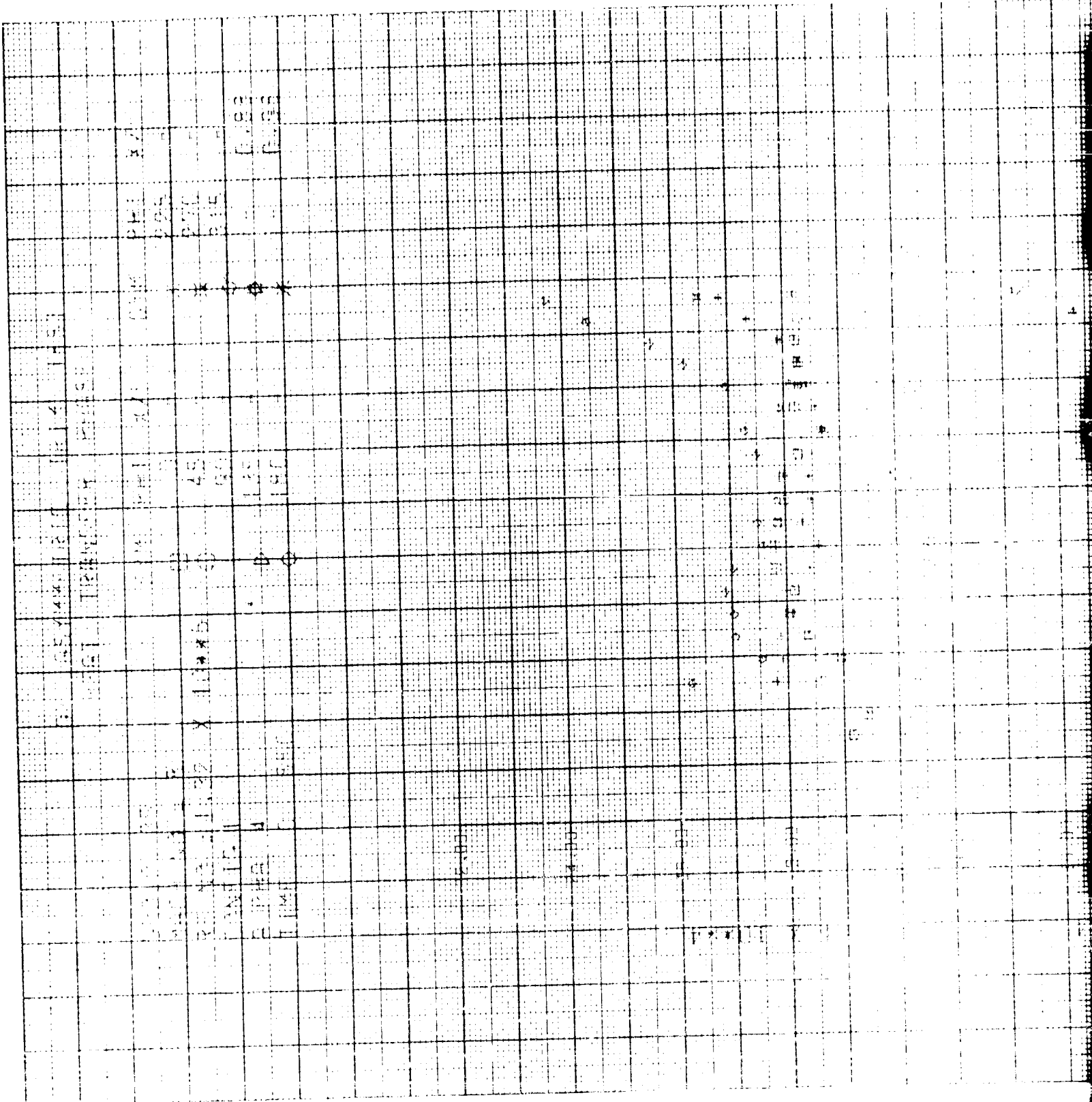
7/14/89

AEUCIAND-INC. J. ANNULO. AFS. TENNESSEE  
VON KARMAN GAS DYNAMICS FACILITY  
50 INCH HYPERSONIC TUNNEL W  
VT0967-M00

GROUP 10 ALPHA-H BETA-H ALPHA-S NULL ALP-P DATE MACH NO TO TIME  
1-IMP P-IMP Q-IMP V-IMP W-IMP X-IMP Y-IMP Z-IMP 7.07 200.10 1200.00 7.00  
9.18000E-01 2.94701E-02 1.11012E-02 1.78110E-03 2.61405E-03 1.55700E-03 1.31917E-03 1.70900E-03 3.14000E-03

Table with columns: TC NO, X/L, PHI, TEMP, SLOPE, Q-STOR, HT, NH, HM, SI. Rows 1-98. Data includes numerical values for various parameters across different test cases.

Large empty grid table on the right side of the page, likely for data recording or analysis.



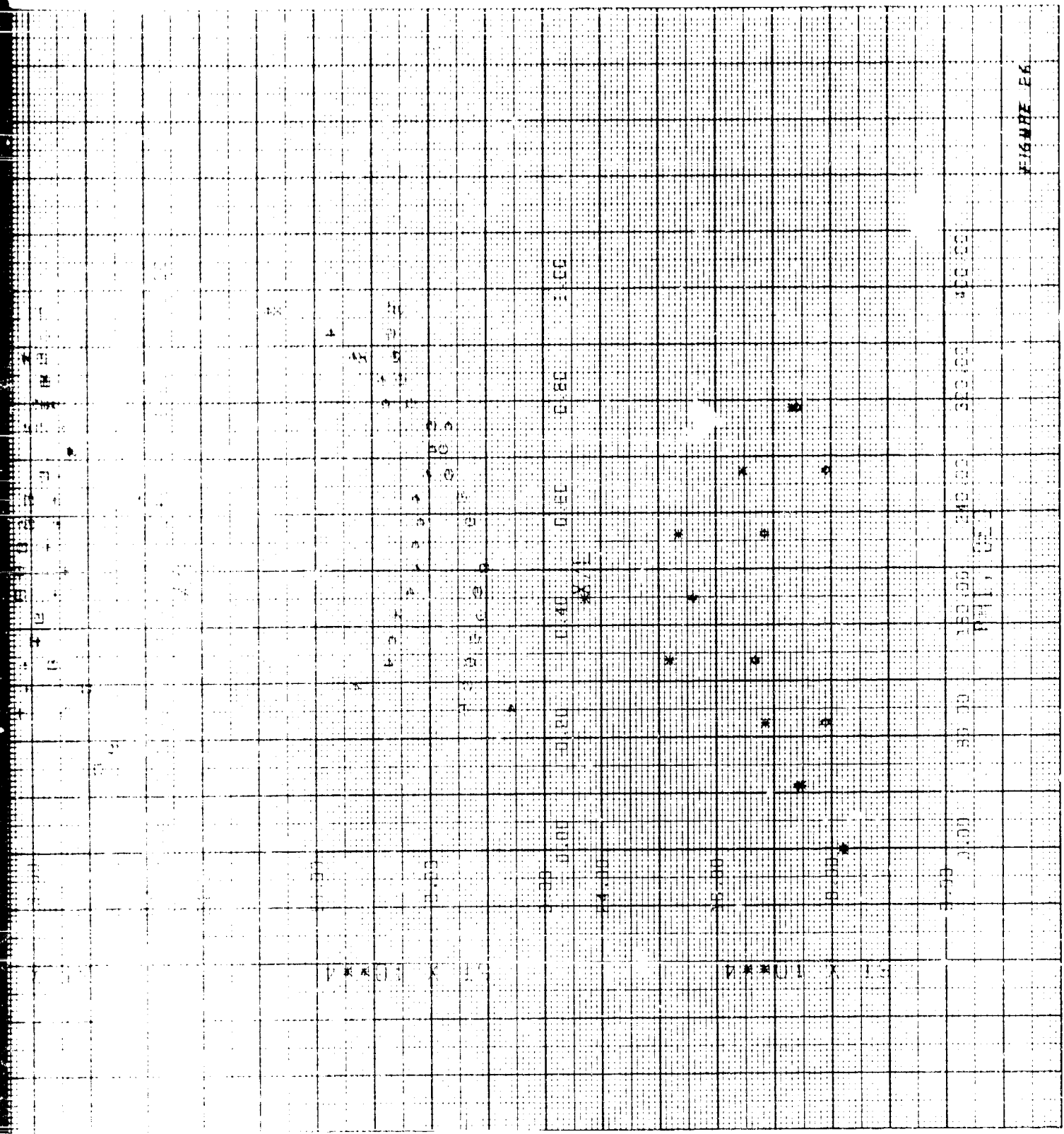


FIGURE E6

Figure 26



DATE	DESCRIPTION	AMOUNT	BALANCE
1950	...	...	...
1951	...	...	...
1952	...	...	...
1953	...	...	...
1954	...	...	...
1955	...	...	...
1956	...	...	...
1957	...	...	...
1958	...	...	...
1959	...	...	...
1960	...	...	...
1961	...	...	...
1962	...	...	...
1963	...	...	...
1964	...	...	...
1965	...	...	...
1966	...	...	...
1967	...	...	...
1968	...	...	...
1969	...	...	...
1970	...	...	...
1971	...	...	...
1972	...	...	...
1973	...	...	...
1974	...	...	...
1975	...	...	...
1976	...	...	...
1977	...	...	...
1978	...	...	...
1979	...	...	...
1980	...	...	...
1981	...	...	...
1982	...	...	...
1983	...	...	...
1984	...	...	...
1985	...	...	...
1986	...	...	...
1987	...	...	...
1988	...	...	...
1989	...	...	...
1990	...	...	...
1991	...	...	...
1992	...	...	...
1993	...	...	...
1994	...	...	...
1995	...	...	...
1996	...	...	...
1997	...	...	...
1998	...	...	...
1999	...	...	...
2000	...	...	...
2001	...	...	...
2002	...	...	...
2003	...	...	...
2004	...	...	...
2005	...	...	...
2006	...	...	...
2007	...	...	...
2008	...	...	...
2009	...	...	...
2010	...	...	...
2011	...	...	...
2012	...	...	...
2013	...	...	...
2014	...	...	...
2015	...	...	...
2016	...	...	...
2017	...	...	...
2018	...	...	...
2019	...	...	...
2020	...	...	...
2021	...	...	...
2022	...	...	...
2023	...	...	...
2024	...	...	...
2025	...	...	...
2026	...	...	...
2027	...	...	...
2028	...	...	...
2029	...	...	...
2030	...	...	...

2

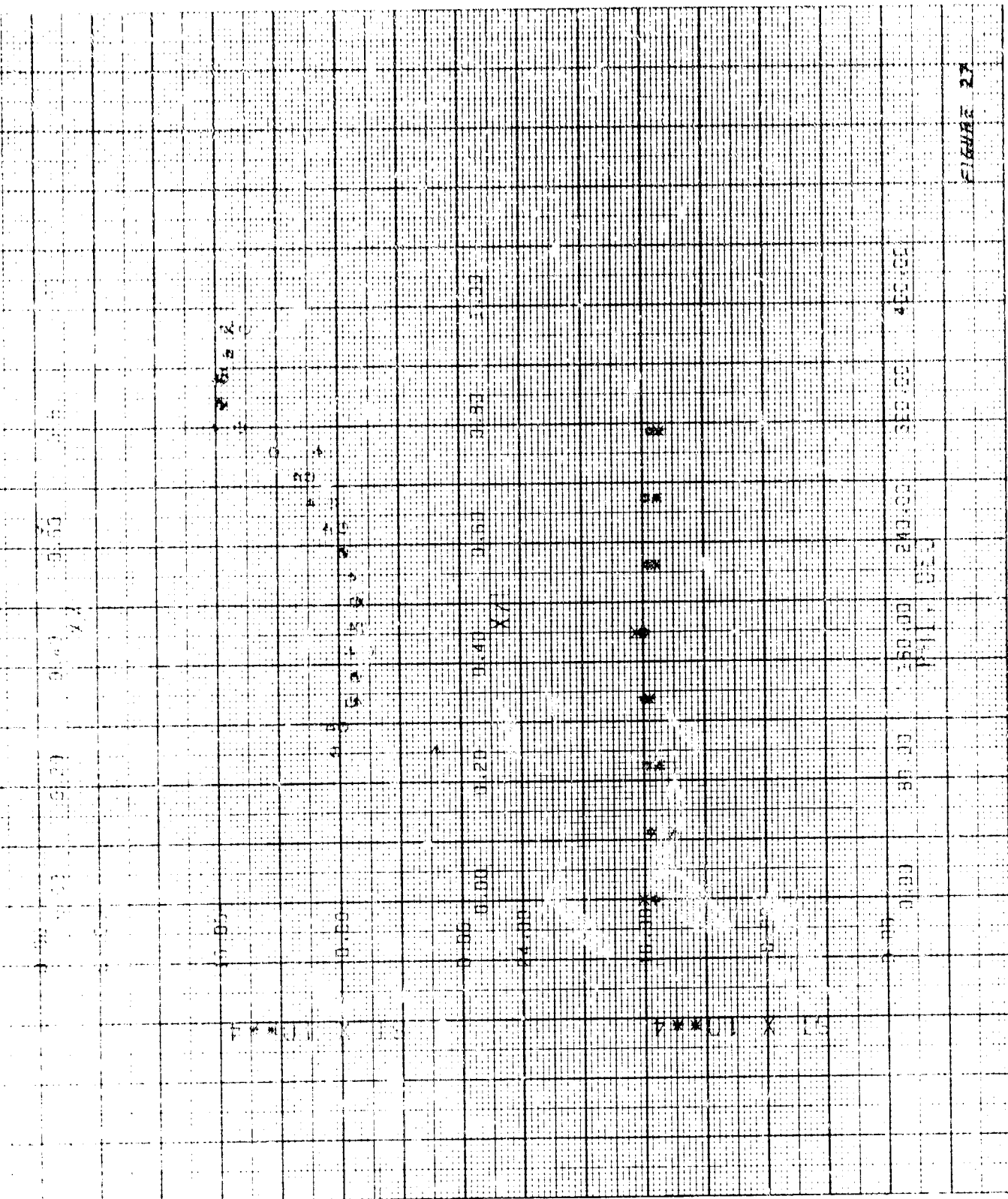


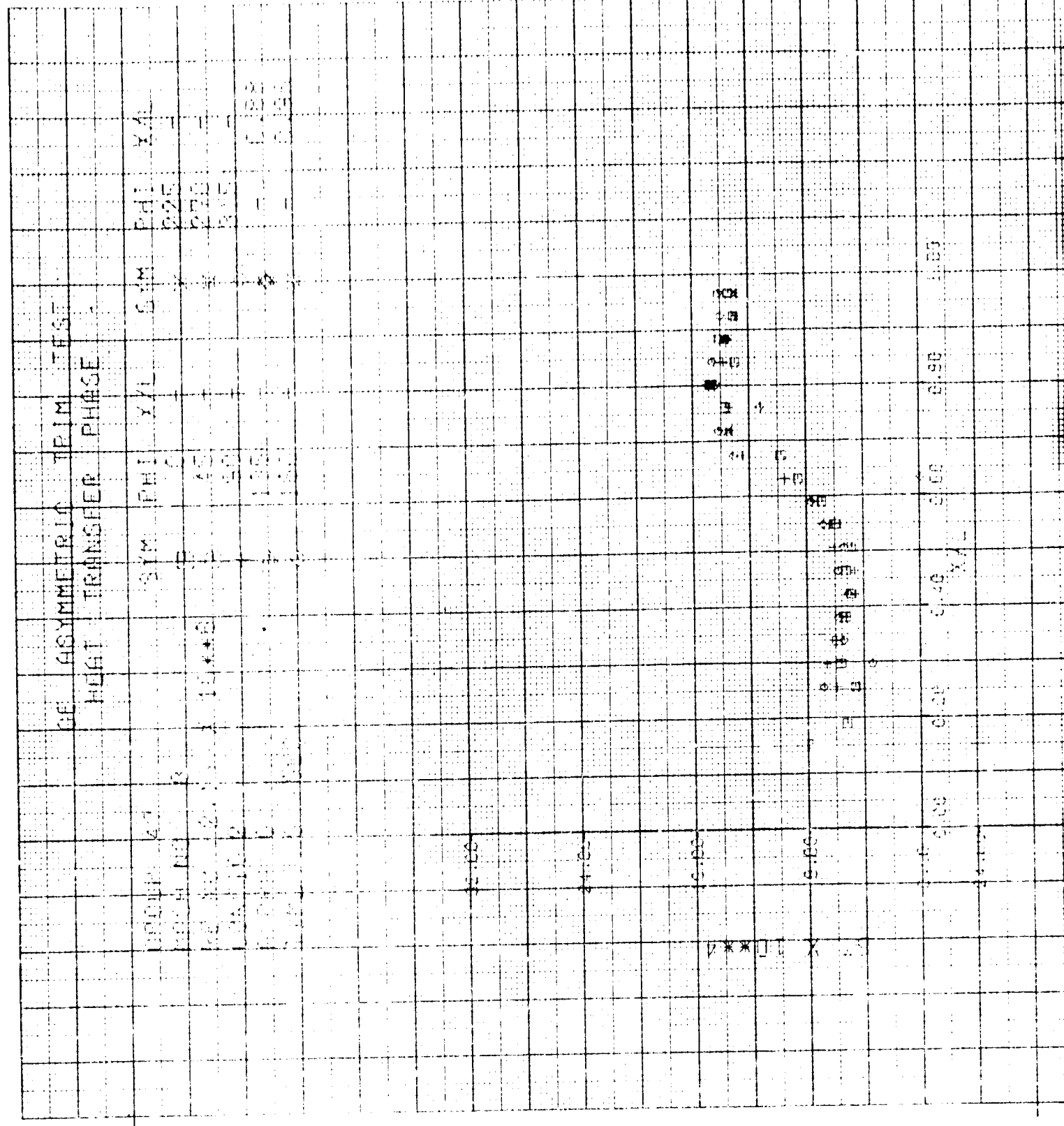
Figure 27

1/16/68

ALCANTARA, INC. ANNOUNCED APPL. TRANSMISSION  
VOM KAMMAN GAS DYNAMICS FACILITY  
50 INCH HYPERSONIC TUNNEL W  
VT1001-000

Table with columns: CHANNEL NO., CHANNEL ID, ALPHA-M, BETA-M, ALPHA-A, ALPHA-A, ALP-P, DATE, MAGN. NO., PPH, TO, TIME. It contains a large grid of numerical data points.

ASYNMETRIC TRIM PHASE  
HULL TRANSFER PHASE  
SYM PHASE



2

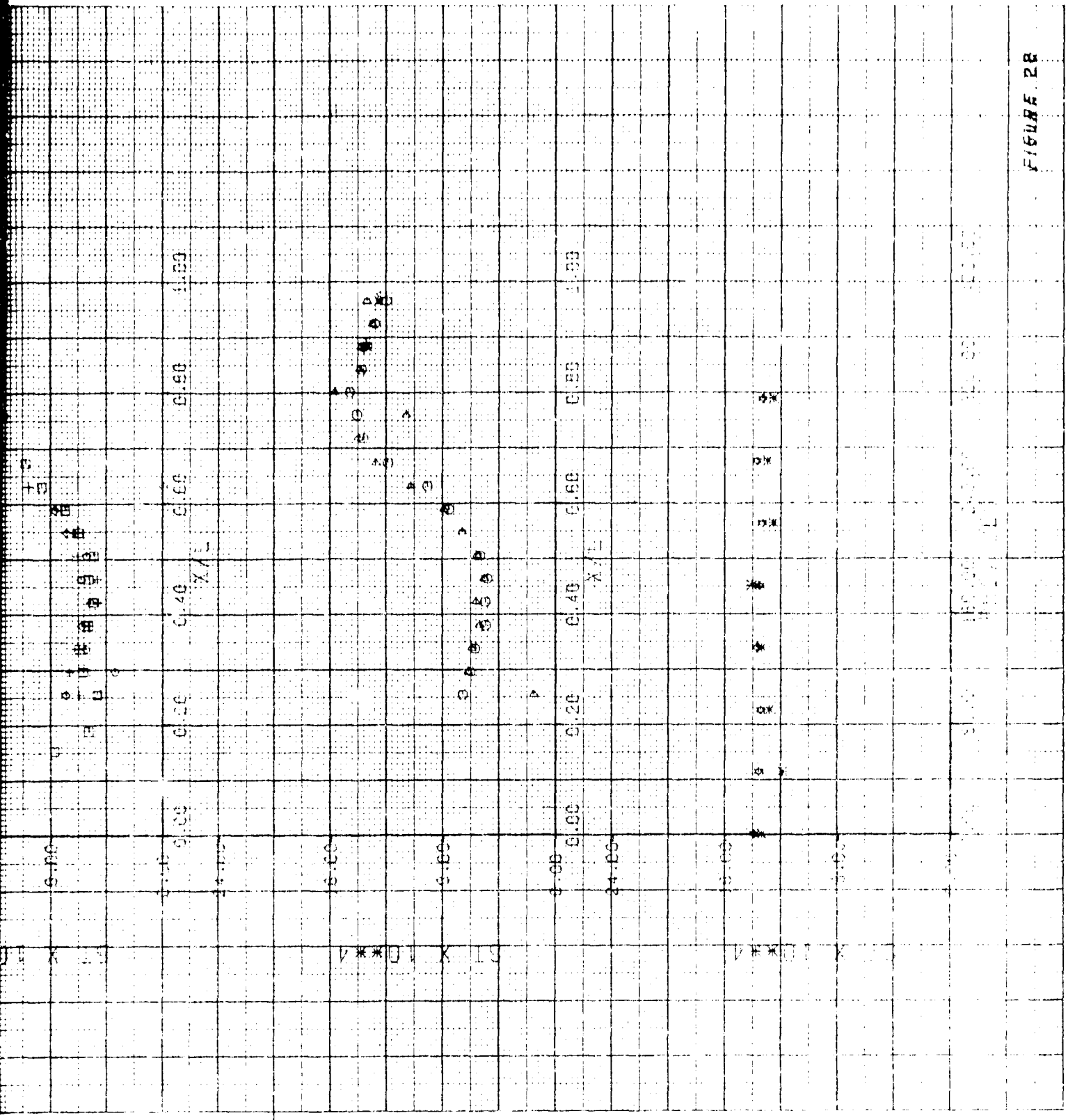


FIGURE 2B

Figure 2B

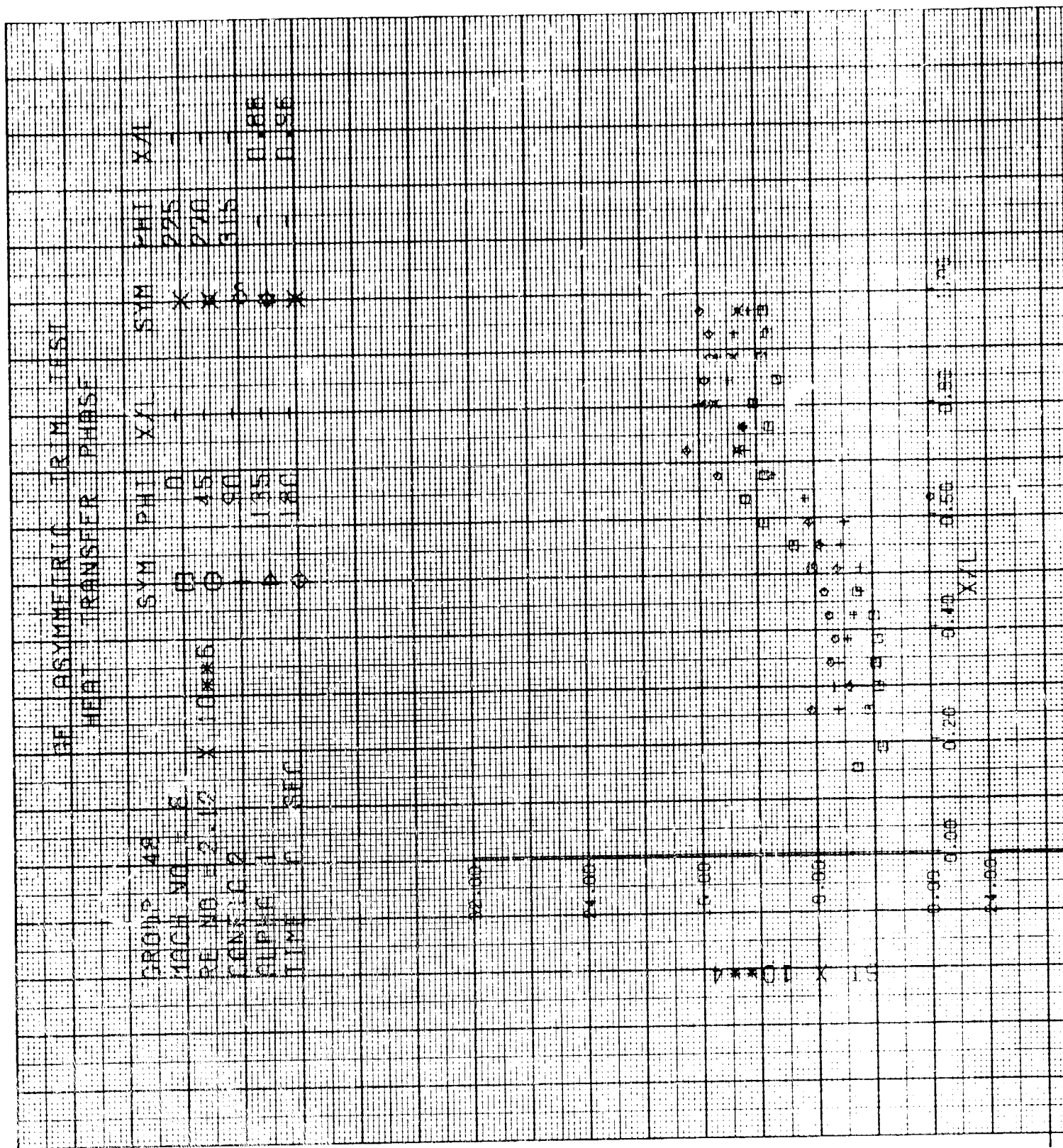
22

7/16/64

ASOCIATED ENGINEERS ANNULAR AIR TENNESSEE  
WIND TUNNEL GAS DYNAMICS FACILITY  
OF THE HYPERSONIC TUNNEL

Table with columns: GROUP, COMPID, ALPHA-M, BETA-M, ALPHA-N, BETA-N, ALPHA-P, BETA-P, DATE, MACH, P1, P2, TIME. It contains a large grid of numerical data points for various test conditions.

ASOCIATED ENGINEERS ANNULAR AIR TENNESSEE  
WIND TUNNEL GAS DYNAMICS FACILITY  
OF THE HYPERSONIC TUNNEL  
GROUP 48



GRAND 48  
 MACH NO 2  
 RE NO = 2.12 x 10<sup>4</sup>  
 GEN 2.2  
 SILPH 1  
 TIME 0 SEC

SYN  
 PHI  
 X/L  
 TRM TEST  
 HEAT TRANSFER PHASE

2

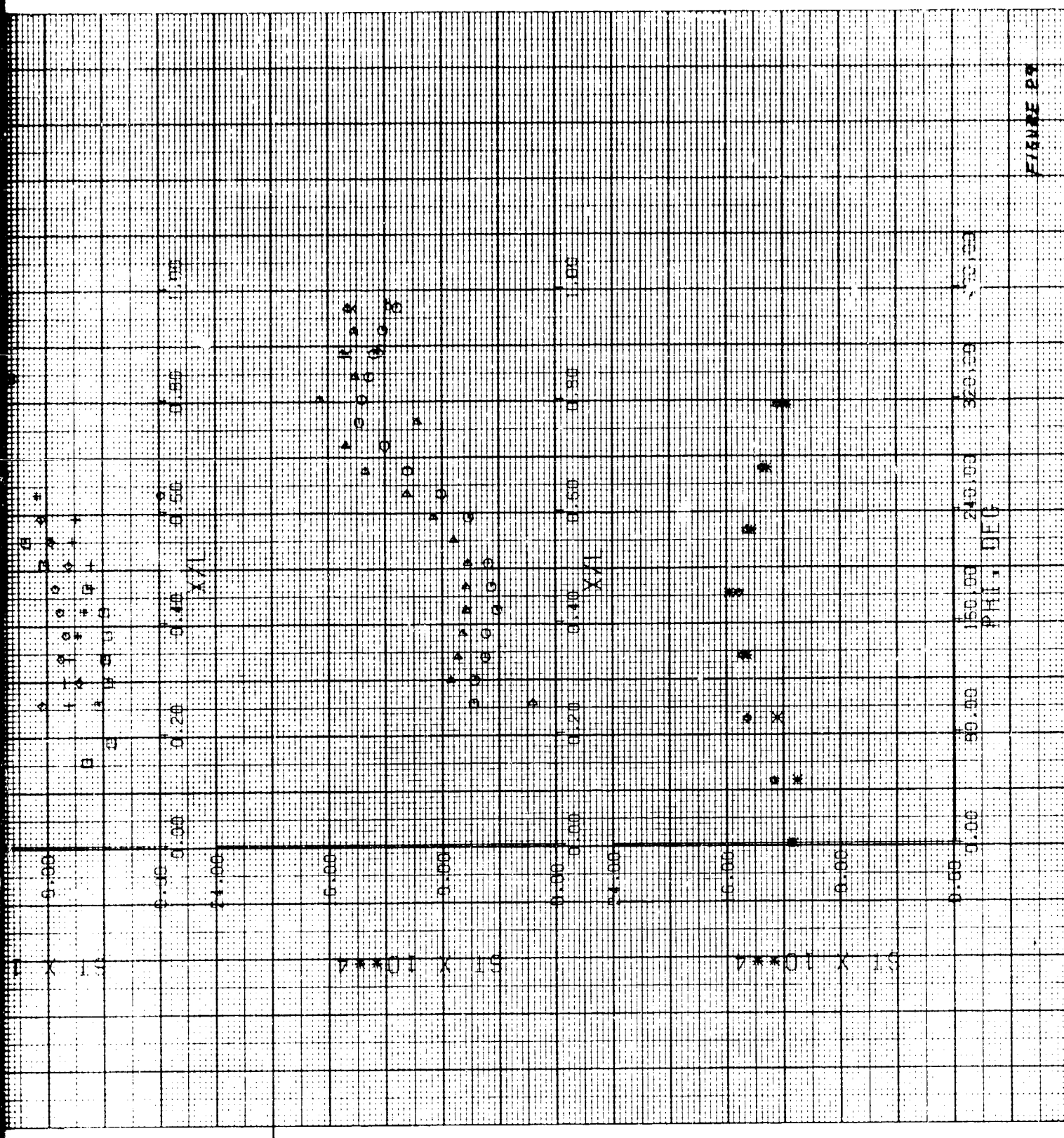


FIGURE 29

Figure 29

7/16/69

MEDICAL (INC) ANNALS OF TENNESSEE  
VON KARMAN GAS DYNAMICS FACILITY  
DE INCORPORATED TUNNEL

GROUP CONT ID ALPHA-H BETA-H ALPHA-H HULL DATE MACH NO PD TO TIME  
48 2 2.00 0 2.00 0 0 7 15 669.67 1305.95  
9.4852E-01 4.85207E-02 2.18434E 3.01303E 0.229175E-05 7.03040E-08 2.12200E 00 1.90400E 00 1.10001E 02

Table with columns: PC NO, X/L, Y/L, TEMP, SLOPE, U-Flow, VY, W, H, M, ST. Rows 1-90 containing numerical data for various parameters.

DE ASYMMETRIC FLOW TEST  
HEAT TRANSFER PHASE  
GROUP 49  
MACH NO 1.8

CELL ASYMMETRIC TRIM TEST  
HEAT TRANSFER PHASE

GROUP 49  
 MACH NO 8  
 RE NO 2-14 X 10\*\*6  
 COND 2  
 ALPHA 2  
 TIME 0 SEC

SYM PHI XYL  
 0 0  
 45 45  
 90 90  
 135 135  
 180 180

SUM PHI XYL  
 X 225  
 X 270  
 0 315  
 0 360  
 X 0

XYL  
 225  
 270  
 315  
 360  
 0

32-00

24-00

6-00

6-00

0-00

12-00

6-00

2\*\*10\*\*X

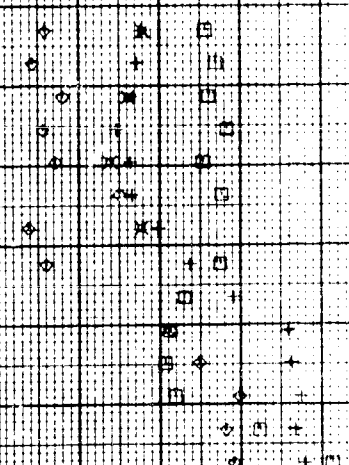
0.45

0.45

XYL

0.45

0.45



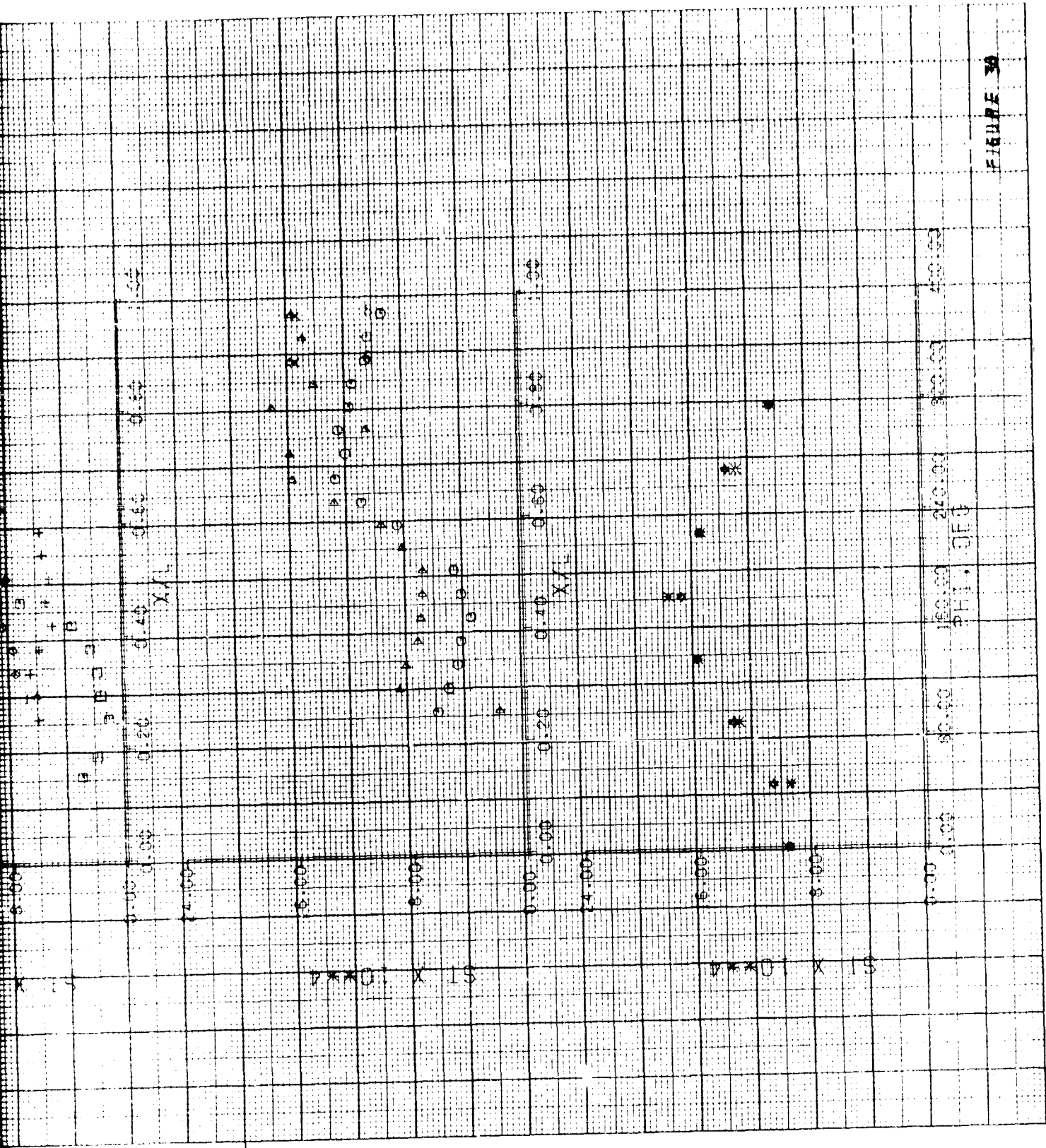


FIGURE 30

Figure 30

7/16/68

ACROBATIC FLIGHT SCHOOL OF CALIFORNIA  
WIND TUNNEL GAS DYNAMICS FACILITY  
50 INCH HYPERSONIC TUNNEL

STATION NO. 01000  
MACH NO. 1.00  
P1 1.00  
P2 1.00  
P3 1.00  
P4 1.00  
P5 1.00  
P6 1.00  
P7 1.00  
P8 1.00  
P9 1.00  
P10 1.00  
P11 1.00  
P12 1.00  
P13 1.00  
P14 1.00  
P15 1.00  
P16 1.00  
P17 1.00  
P18 1.00  
P19 1.00  
P20 1.00  
P21 1.00  
P22 1.00  
P23 1.00  
P24 1.00  
P25 1.00  
P26 1.00  
P27 1.00  
P28 1.00  
P29 1.00  
P30 1.00  
P31 1.00  
P32 1.00  
P33 1.00  
P34 1.00  
P35 1.00  
P36 1.00  
P37 1.00  
P38 1.00  
P39 1.00  
P40 1.00  
P41 1.00  
P42 1.00  
P43 1.00  
P44 1.00  
P45 1.00  
P46 1.00  
P47 1.00  
P48 1.00  
P49 1.00  
P50 1.00  
P51 1.00  
P52 1.00  
P53 1.00  
P54 1.00  
P55 1.00  
P56 1.00  
P57 1.00  
P58 1.00  
P59 1.00  
P60 1.00  
P61 1.00  
P62 1.00  
P63 1.00  
P64 1.00  
P65 1.00  
P66 1.00  
P67 1.00  
P68 1.00  
P69 1.00  
P70 1.00  
P71 1.00  
P72 1.00  
P73 1.00  
P74 1.00  
P75 1.00  
P76 1.00  
P77 1.00  
P78 1.00  
P79 1.00  
P80 1.00  
P81 1.00  
P82 1.00  
P83 1.00  
P84 1.00  
P85 1.00  
P86 1.00  
P87 1.00  
P88 1.00  
P89 1.00  
P90 1.00  
P91 1.00  
P92 1.00  
P93 1.00  
P94 1.00  
P95 1.00  
P96 1.00  
P97 1.00  
P98 1.00  
P99 1.00  
P100 1.00

Table with columns: P1, P2, P3, P4, P5, P6, P7, P8, P9, P10, P11, P12, P13, P14, P15, P16, P17, P18, P19, P20, P21, P22, P23, P24, P25, P26, P27, P28, P29, P30, P31, P32, P33, P34, P35, P36, P37, P38, P39, P40, P41, P42, P43, P44, P45, P46, P47, P48, P49, P50, P51, P52, P53, P54, P55, P56, P57, P58, P59, P60, P61, P62, P63, P64, P65, P66, P67, P68, P69, P70, P71, P72, P73, P74, P75, P76, P77, P78, P79, P80, P81, P82, P83, P84, P85, P86, P87, P88, P89, P90, P91, P92, P93, P94, P95, P96, P97, P98, P99, P100. Each row contains numerical data for each pressure point.

Large empty table with multiple columns and rows, possibly for data recording or analysis.

GE	ASYMMETRIC	TRIM	TEST
	HECI	TRANSFER	
GROUP 50	SYM	PHI	EIM
GCH NO 1 B		PHI	X2
RE NO 2-18 X 10*15	15	45	PDS
CNTIC 2	G	GC	PDS
FLIPER 4	0	135	G15
FLIPER 0 SEC	0	180	PDS
			PDS
11-00			
14-00		R	A
15-00	0	R	A
16-00	0	R	A
17-00	+	R	A
18-00	+	R	A
19-00	+	R	A
20-00	+	R	A
21-00	+	R	A
22-00	+	R	A
23-00	+	R	A
24-00	+	R	A

11

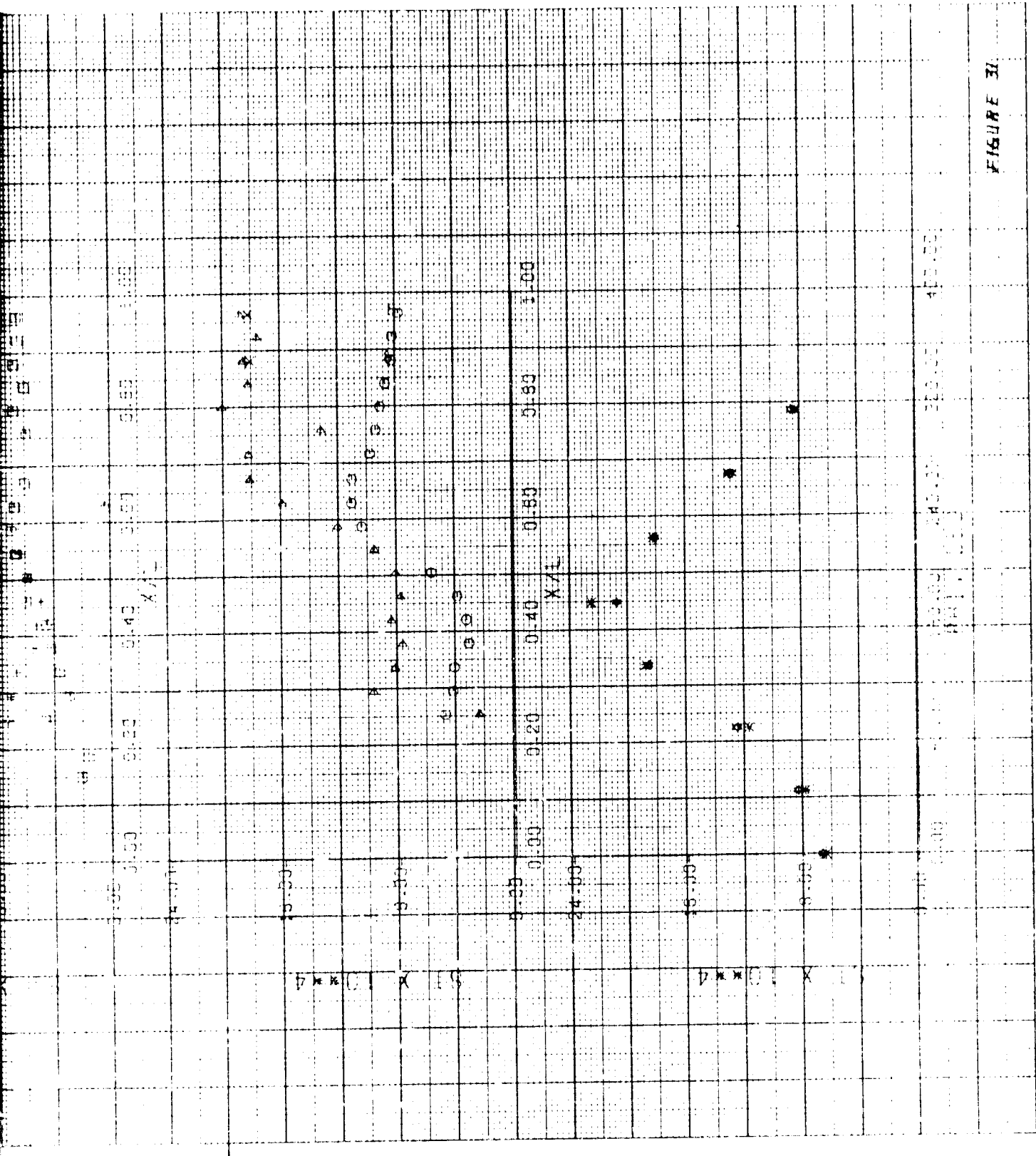


FIGURE 30

Figure 31

111

7/16/69

MEDICAL INC. - ANNUAL REPORT FOR THE YEAR 1968  
MEMPHIS MASS DYNAMICS FACILITY  
90 INCH HYPERSONIC TUNNEL  
VT0047-000

Table with columns: GROUP, COMP ID, ALPHA-H, BETA-H, ALPHA-H, BETA-H, DATE, MACH NO, TO, TIME. Rows contain numerical data for various test runs.

Table with columns: GROUP, COMP ID, ALPHA-H, BETA-H, ALPHA-H, BETA-H, DATE, MACH NO, TO, TIME. Rows contain numerical data for various test runs.

12

GROUP	ASYMMETRIC	TRIM	TEST	PHI	PHI	PHI	PHI	PHI	PHI
MACH NO	HEAT	TRANSFER	PHASE	SYM	PHI	X1	SYM	PHI	X2
51									
8									
2.18 X 10**6									
2									
3									
0									
32.00									
34.00									
36.00									
38.00									
40.00									
42.00									
44.00									
46.00									
48.00									
50.00									
52.00									
54.00									
56.00									
58.00									
60.00									
62.00									
64.00									
66.00									
68.00									
70.00									
72.00									
74.00									
76.00									
78.00									
80.00									
82.00									
84.00									
86.00									
88.00									
90.00									
92.00									
94.00									
96.00									
98.00									
100.00									

51 X 10\*\*4

54

GROUP 51  
 MACH NO 8  
 RE NO 2.18 X 10\*\*6  
 CNFIC 2  
 ALPHA 3  
 TIME 0 SEC

32.00  
 34.00  
 36.00  
 38.00  
 40.00  
 42.00  
 44.00  
 46.00  
 48.00  
 50.00  
 52.00  
 54.00  
 56.00  
 58.00  
 60.00  
 62.00  
 64.00  
 66.00  
 68.00  
 70.00  
 72.00  
 74.00  
 76.00  
 78.00  
 80.00  
 82.00  
 84.00  
 86.00  
 88.00  
 90.00  
 92.00  
 94.00  
 96.00  
 98.00  
 100.00

PH1 225  
 PH1 272  
 PH1 335  
 PH1 400  
 PH1 465  
 PH1 530

X1 0  
 X1 45  
 X1 90  
 X1 135  
 X1 180

SYM \*  
 SYM \*  
 SYM \*  
 SYM \*  
 SYM \*

PH1 225  
 PH1 272  
 PH1 335  
 PH1 400  
 PH1 465  
 PH1 530

X1 0  
 X1 45  
 X1 90  
 X1 135  
 X1 180

SYM \*  
 SYM \*  
 SYM \*  
 SYM \*  
 SYM \*

PH1 225  
 PH1 272  
 PH1 335  
 PH1 400  
 PH1 465  
 PH1 530

51 X 10\*\*4  
 54

54

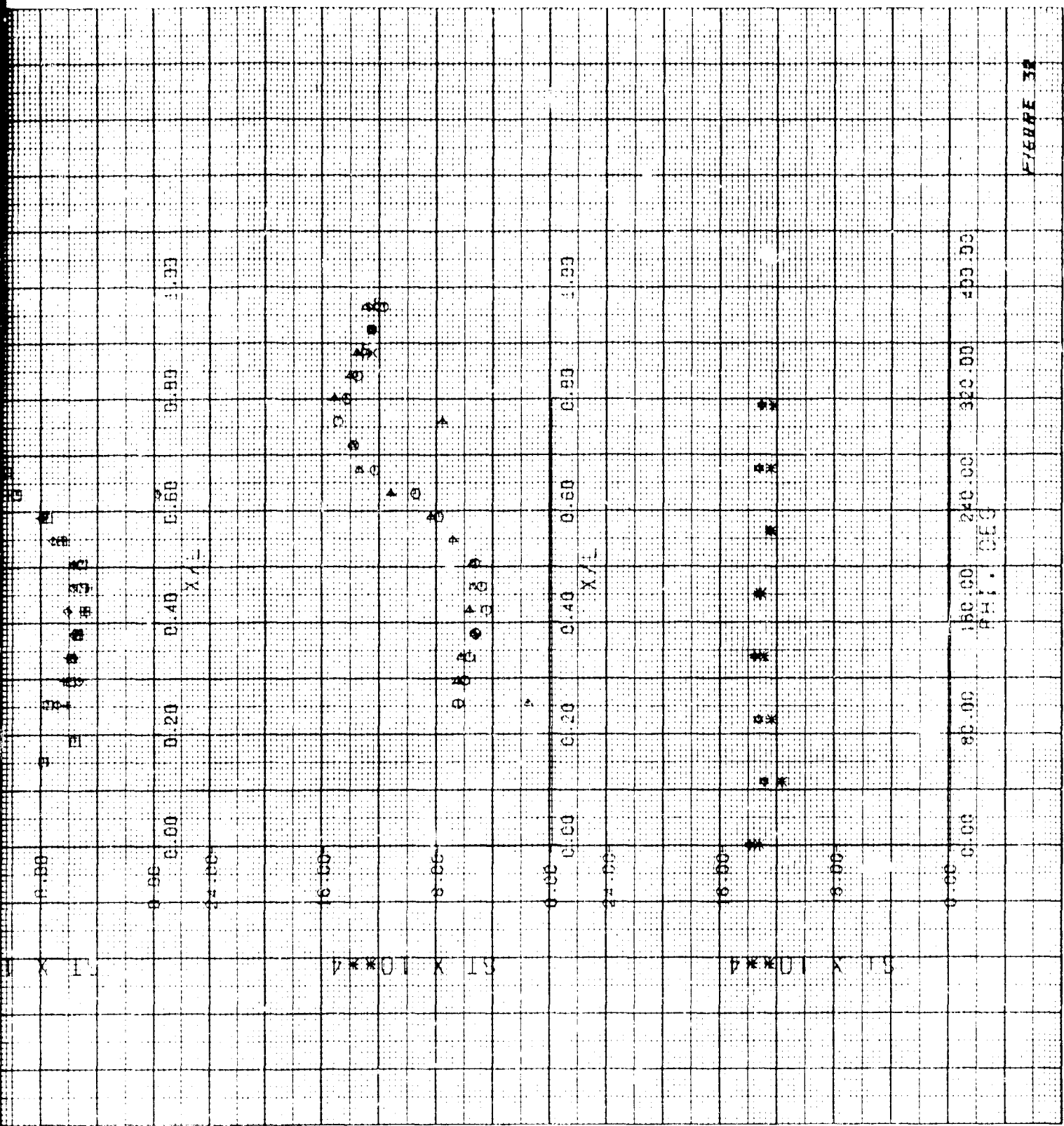


FIGURE 32

Figure 32

11/53



DEPARTMENT OF AGRICULTURE

HEAT TRANSFER PAPER

STATIONARY STATE

HEAT TRANSFER

STATIONARY STATE

HEAT TRANSFER

STATIONARY STATE

HEAT TRANSFER

STATIONARY STATE

HEAT TRANSFER

STATIONARY STATE

HEAT TRANSFER

STATIONARY STATE

HEAT TRANSFER

STATIONARY STATE

HEAT TRANSFER

STATIONARY STATE

HEAT TRANSFER

STATIONARY STATE

HEAT TRANSFER

STATIONARY STATE

HEAT TRANSFER

STATIONARY STATE

DATE

NO. 1

1920

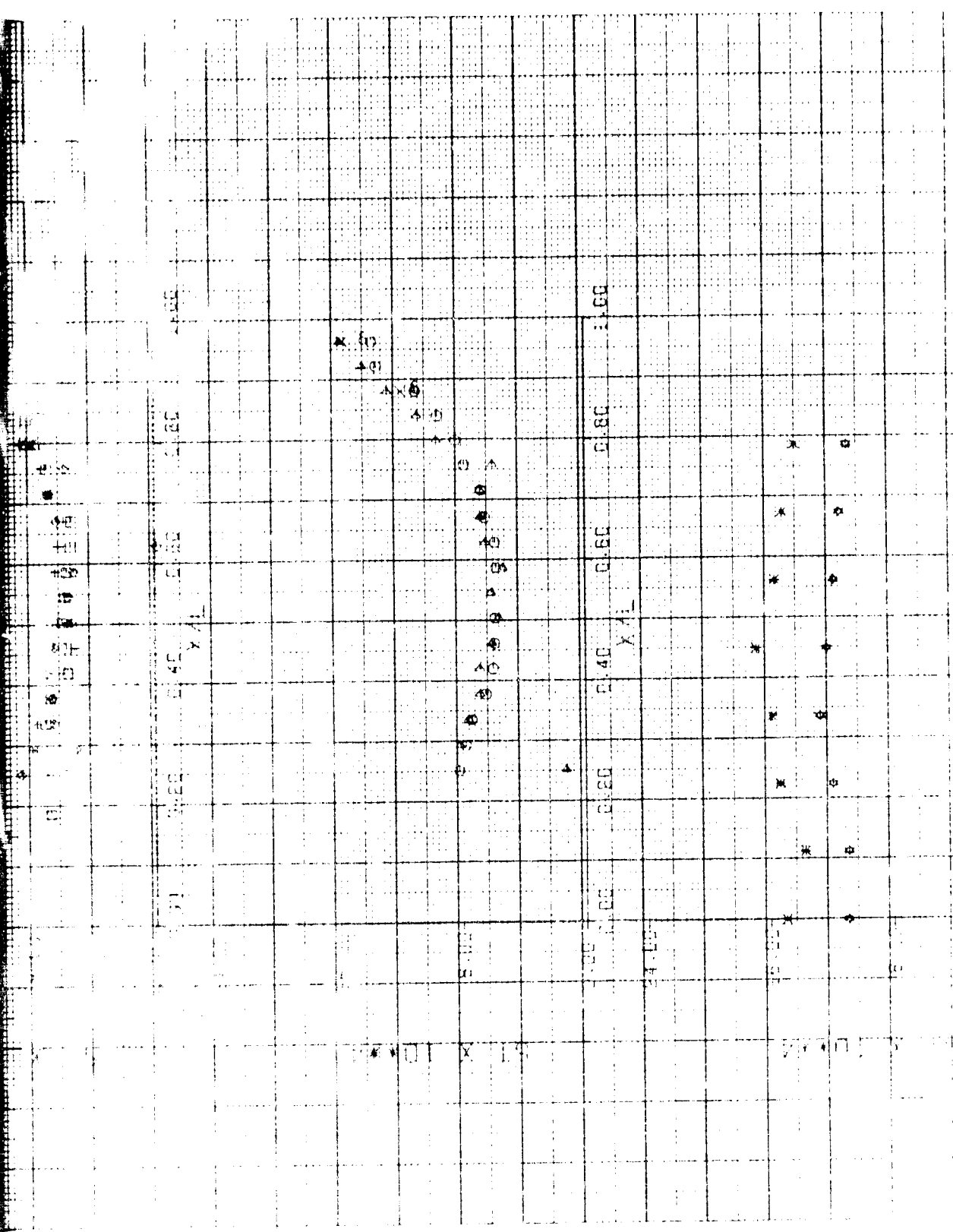


Figure 1

3

7/16/60

AMDC (AMU, INC.) ANNUL AND TUNNEL  
VON KARMAN GAS DYNAMICS FACILITY  
30 INCH HYPERSONIC TUNNEL  
W-10567-SUO

ORIGIN: CON-10 2 ALPHA-M 1-03 BETA-N 0 ALPHA-S 1-07 MULL 0 G-PLP 0 DATE 7/5  
FACH NO 7.07 PO 281.00 TO 1841.01 TIME 6.00  
REV# ME/0  
REVISIONS: 02 1-3159E-02 1-3159E-00 3.7001E-03 2.0000E-05 7.0000E-05 7.0000E-05  
1-2000E-00 1-2000E-00 3.0011E-02

Table with columns: TL, NO, X, Y, PHI, TEMP, SLOPE, Q-STON, HT, MM, PSI, ST. Contains numerical data for various points in the tunnel.

SYMMETRIC TUNNEL TEST  
HEAT TRANSFER PHENOMENON  
SYMM. PHI X  
SYMM. PHI Y  
SYMM. PHI Z

7/16

GROUP	ASYMMETRIC		IRIM		TEST		
	HTGT	TRANSFER	PHI	Y/V	SYM	PHI	Y/V
43					*	225	-
NO. 1-8			45		*	270	-
NO. 10-18			90		*	315	-
NO. 1			135		*	-	BLER
NO. 1			180		*	-	BLER
NO. 2							
NO. 3							
NO. 4							
NO. 5							
NO. 6							
NO. 7							
NO. 8							
NO. 9							
NO. 10							
NO. 11							
NO. 12							
NO. 13							
NO. 14							
NO. 15							
NO. 16							
NO. 17							
NO. 18							
NO. 19							
NO. 20							
NO. 21							
NO. 22							
NO. 23							
NO. 24							
NO. 25							
NO. 26							
NO. 27							
NO. 28							
NO. 29							
NO. 30							
NO. 31							
NO. 32							
NO. 33							
NO. 34							
NO. 35							
NO. 36							
NO. 37							
NO. 38							
NO. 39							
NO. 40							
NO. 41							
NO. 42							
NO. 43							
NO. 44							
NO. 45							
NO. 46							
NO. 47							
NO. 48							
NO. 49							
NO. 50							

1

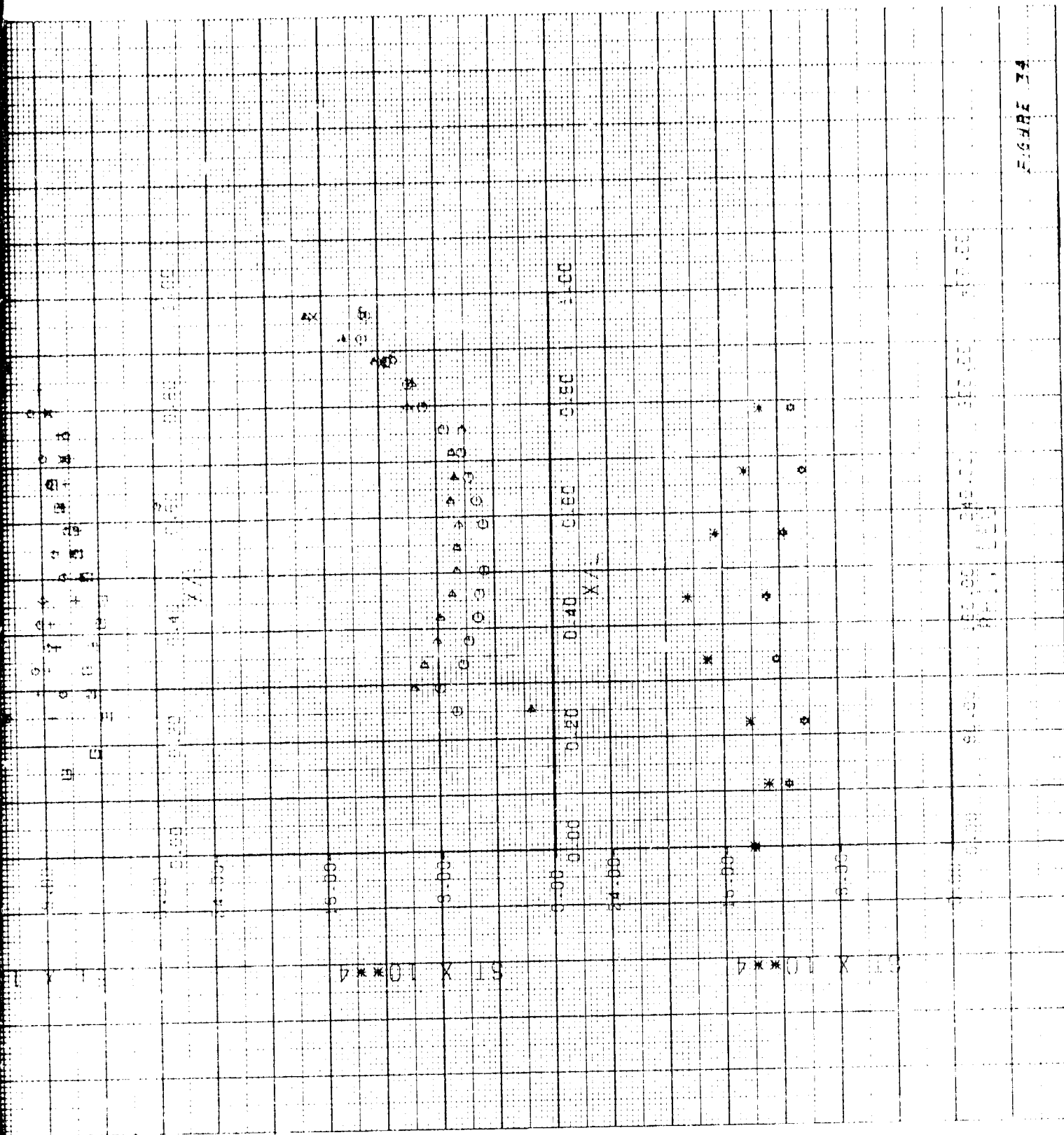
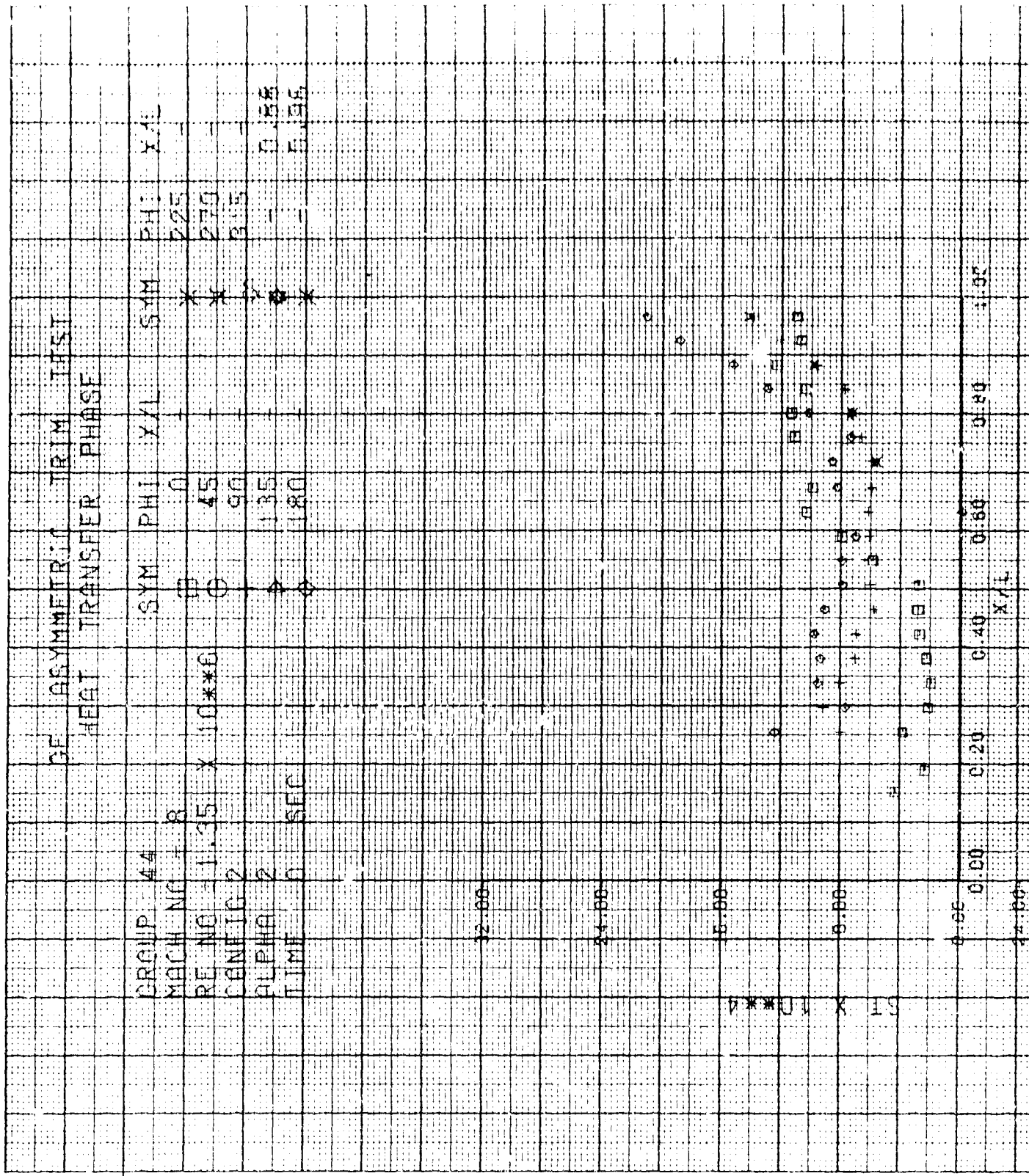


FIGURE 34

Figure 34





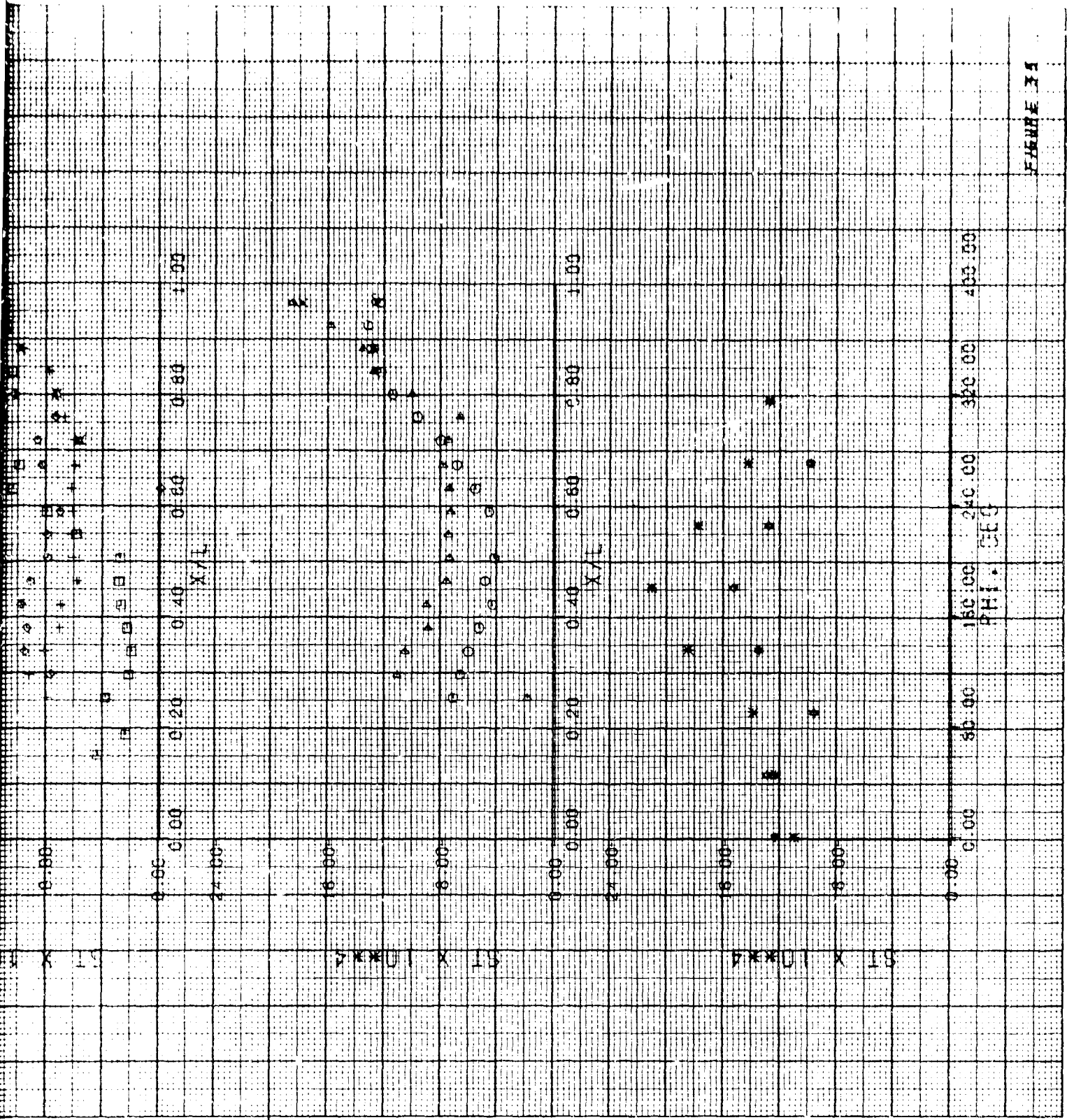
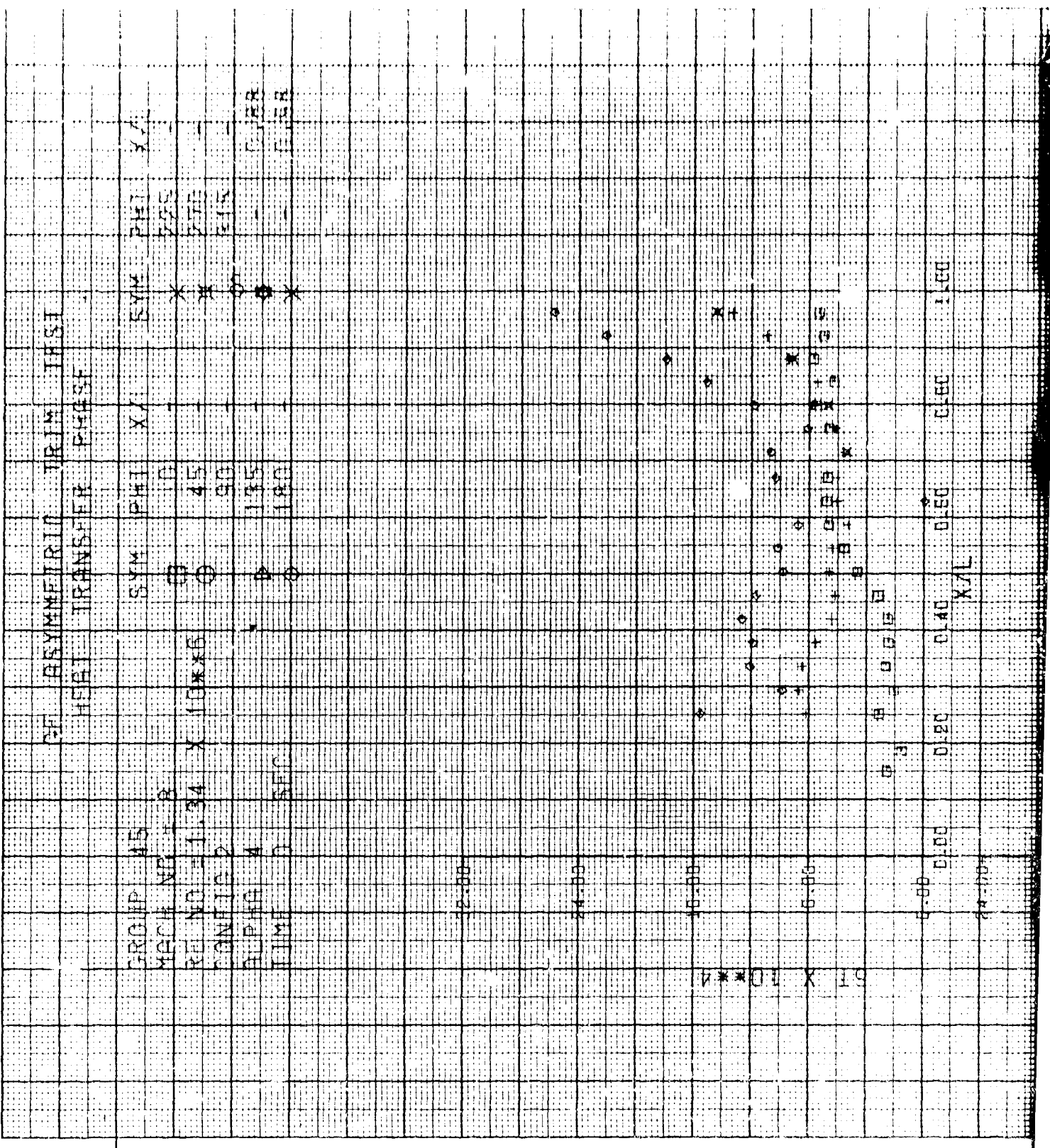


PLATE 34

Figure 35

3





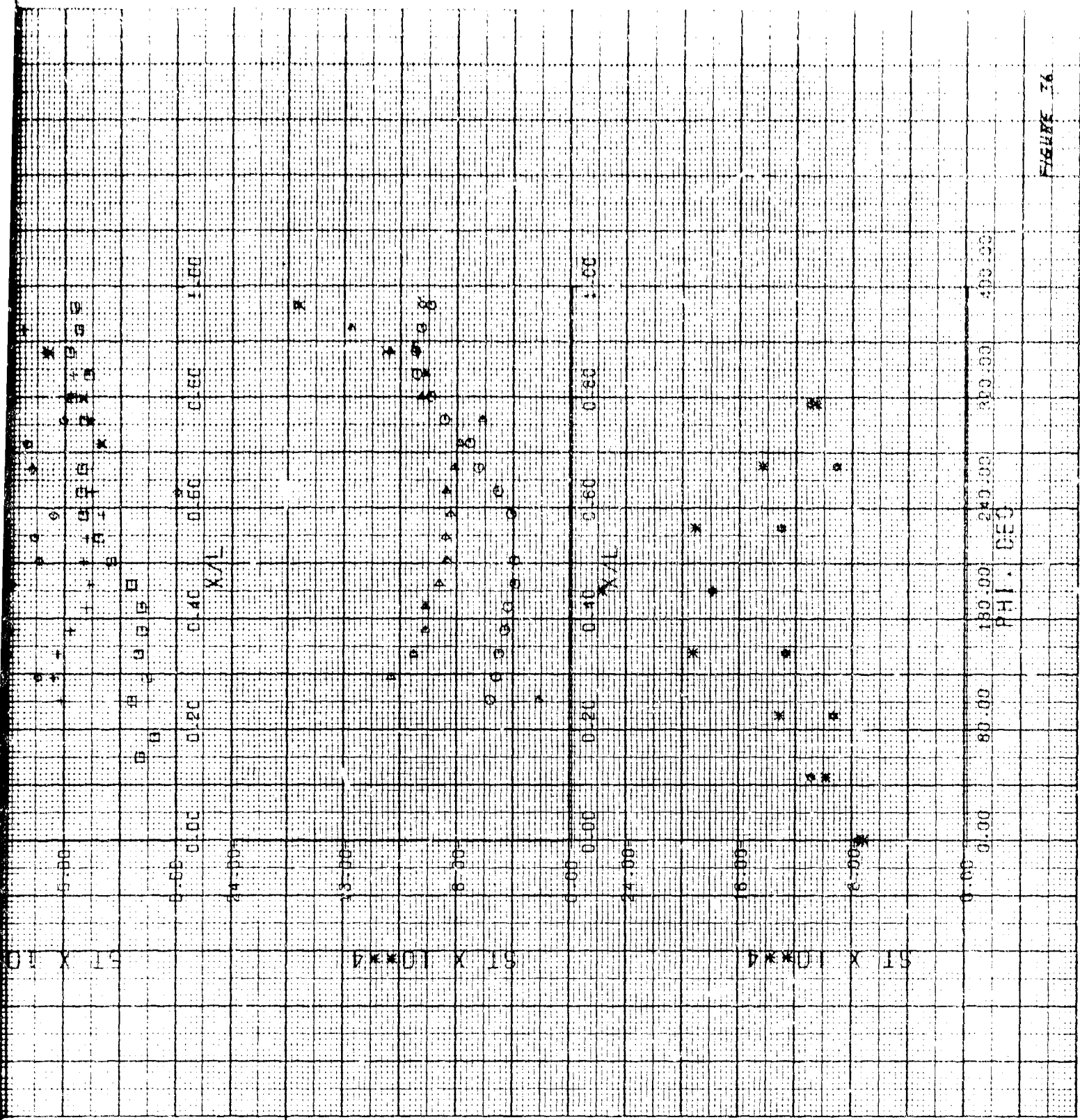


Figure 36

FIGURE 36

55

7/14/49

MEMORANDUM FOR THE RECORD  
SUBJECT: [Illegible]

Table with columns for ALPHABETIC, ALPHABETIC, ALPHABETIC, ALPHABETIC, ALPHABETIC, ALPHABETIC, ALPHABETIC, ALPHABETIC, ALPHABETIC. Rows 1-98.

Vertical text on the right side, possibly a list of names or identifiers, arranged in columns.

11.00  
11.00

BE ASYMMETRIC FILM TEST  
HEAT TRANSFER PHASE

GROUP	AS	SYM	PHI	XA	SUM	PHI	XA
MREH	NE	+	5			208	
RE	50	+	1.33	+	45	274	
CONF	1.0					3.5	
PLANE	1						
TIME							

32.00

24.00

16.00

BT X 11\*\*4

8.00

6.00

4.00

TIME	PHI	XA
0.00	0.40	0.80
0.50	0.60	1.00

2

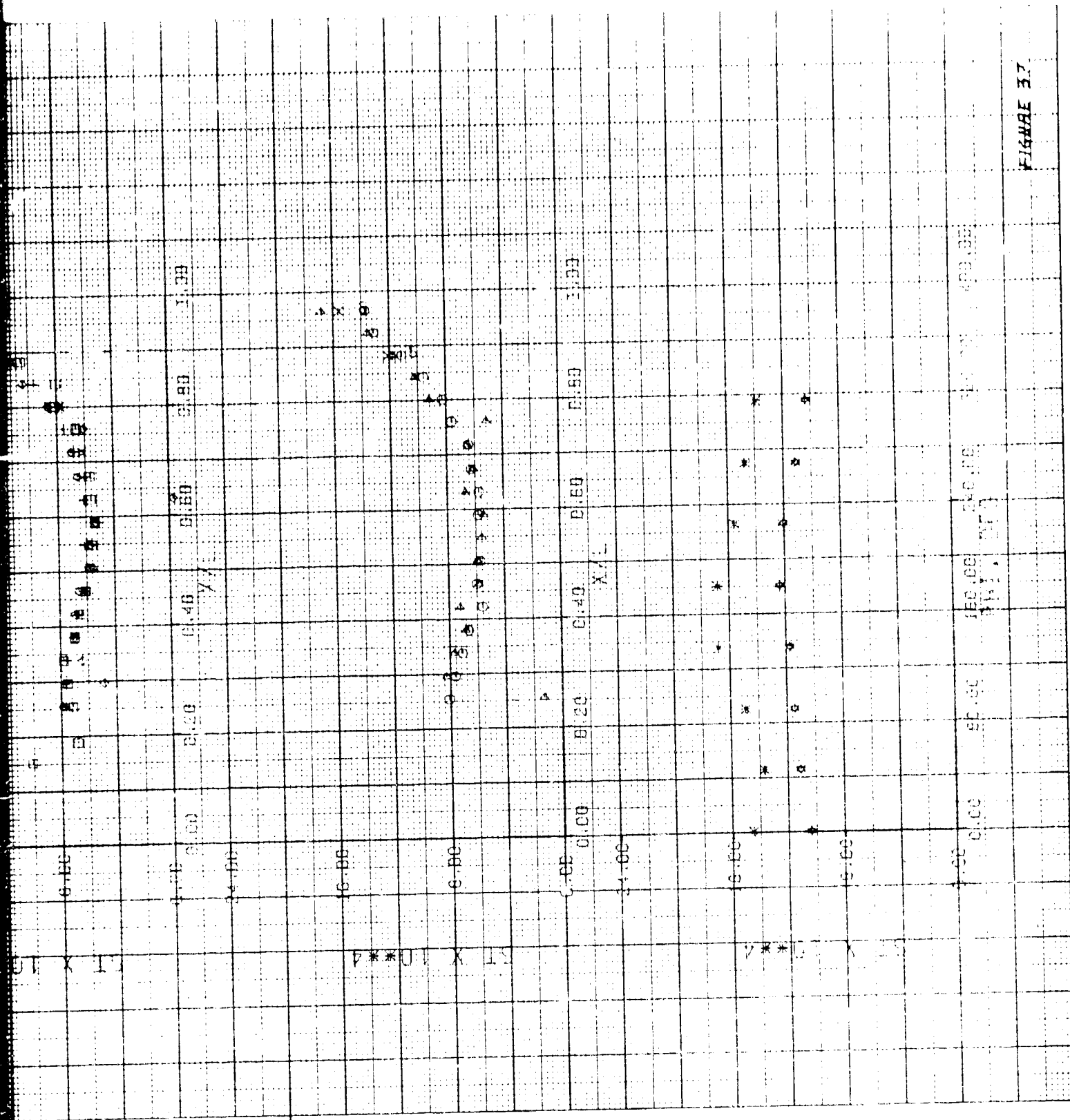


FIGURE 37

Figure 37

3

7/14/69

MECHANICAL ANALYSIS OF THE TENSILE  
AND BUCKLING BEHAVIOR OF A  
CIRCULAR CRYSTALLINE POLYMER

GROUP NO. 101 ALPHA-1 0.010000 BETA-1 0.010000 ALPHA-2 0.010000 BETA-2 0.010000 ALPHA-3 0.010000 BETA-3 0.010000 ALPHA-4 0.010000 BETA-4 0.010000 ALPHA-5 0.010000 BETA-5 0.010000 ALPHA-6 0.010000 BETA-6 0.010000 ALPHA-7 0.010000 BETA-7 0.010000 ALPHA-8 0.010000 BETA-8 0.010000 ALPHA-9 0.010000 BETA-9 0.010000 ALPHA-10 0.010000 BETA-10 0.010000 ALPHA-11 0.010000 BETA-11 0.010000 ALPHA-12 0.010000 BETA-12 0.010000 ALPHA-13 0.010000 BETA-13 0.010000 ALPHA-14 0.010000 BETA-14 0.010000 ALPHA-15 0.010000 BETA-15 0.010000 ALPHA-16 0.010000 BETA-16 0.010000 ALPHA-17 0.010000 BETA-17 0.010000 ALPHA-18 0.010000 BETA-18 0.010000 ALPHA-19 0.010000 BETA-19 0.010000 ALPHA-20 0.010000 BETA-20 0.010000

Table with columns: NO, PNI, TENS, STRESS, STRAIN, etc. Rows 1-90.

OF SYMMETRIC TENSILE  
HEAT TRANSFER BEHAVOR

GROUP 1

GROUP

GE	ASYMMETRIC TRIM TEST											
	HGT	TRANSFER PHASE										
GROUP	S	SYM	PHI	X	Y	Z	SYM	PHI	X	Y	Z	SYM
MACH NO	1.8											
RE XC	0.179	Y	0	AS	AS	AS	*	PHI				
CONFIG				EC	EC	EC	*	PHI				
PARMS				135	135	135	*	PHI				
TIME				180	180	180	*	PHI				
09:00												
09:40												
09:01												
09:00												

2

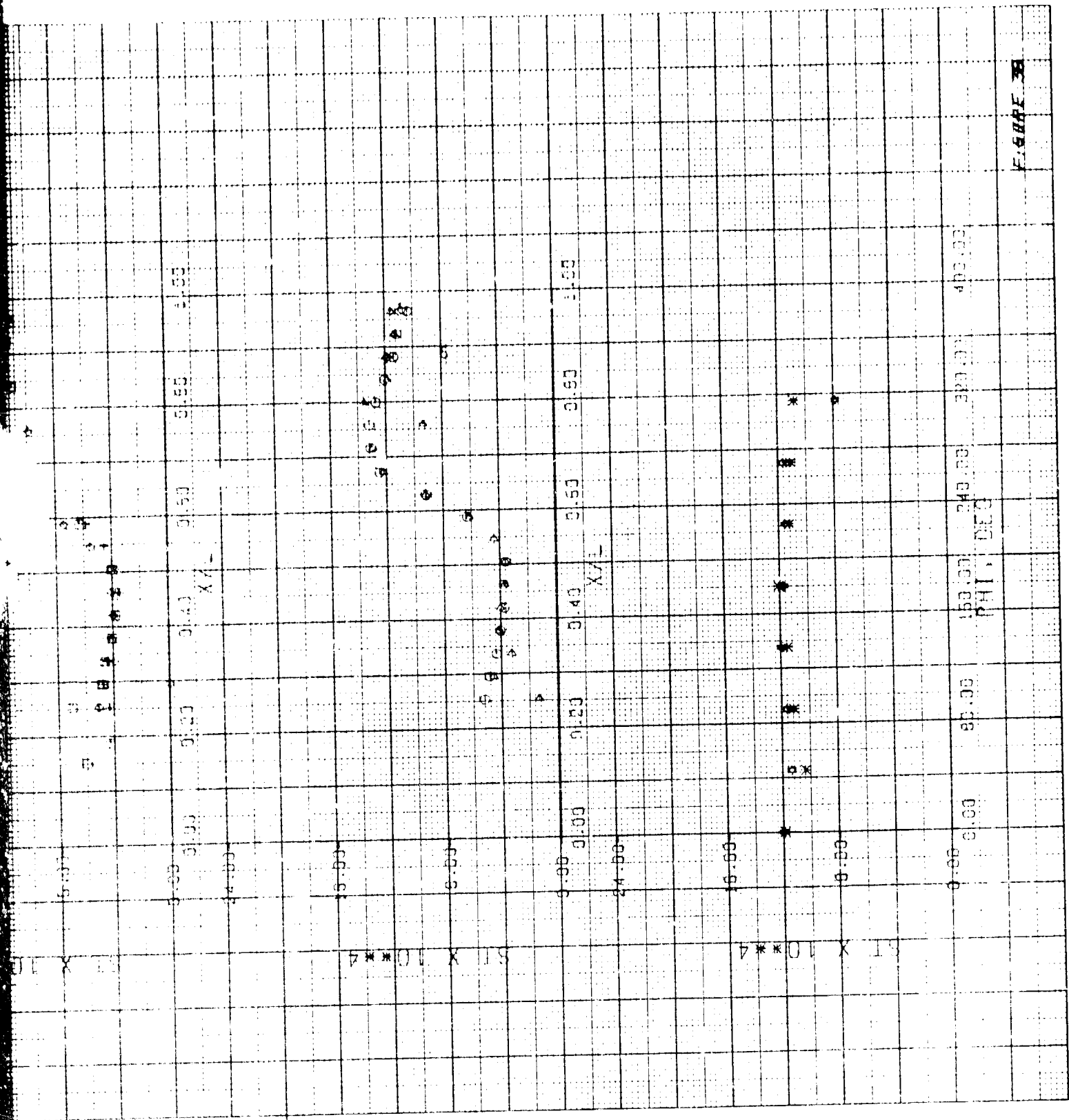


Figure 38

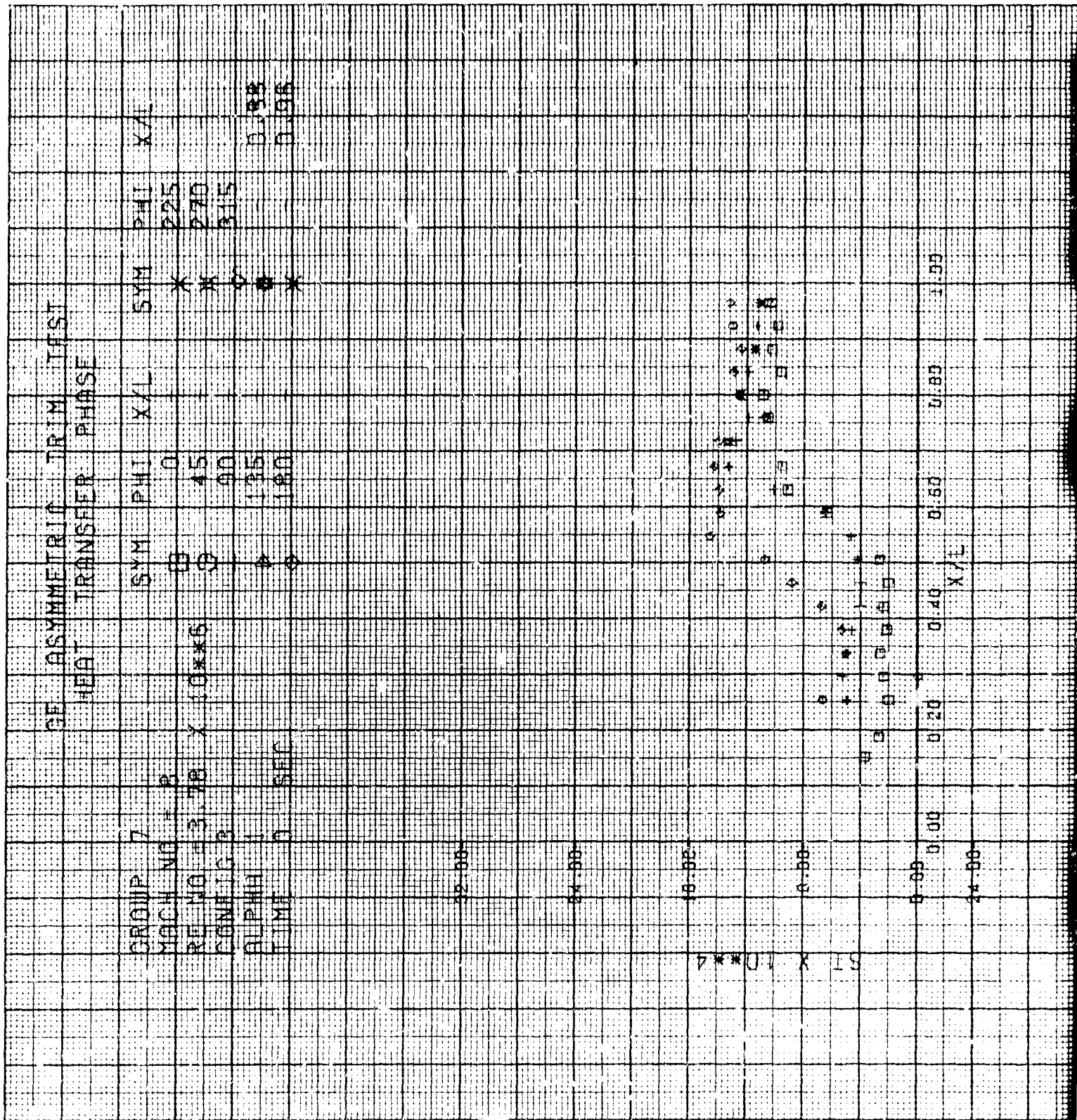
3

7/16/60

MEDICAL INC. 1 ANNUL AFS, TENNESSEE  
VON KARMAN GAS DYNAMICS FACILITY  
80 INCH HYPERSONIC TUNNEL #  
V10947-000

Table with columns: GROUP, COMP ID, ALPHA-H, BETA-H, ALPHA-H, ROLL, ALPHA-H, DATE, MACH NO, PC, TU, TIME. Rows include data for various test points and conditions.

SYMMETRIC TRIM TEST  
HEAT TRANSFER PHASE  
SYMM PHIL X/L  
GROUP 7



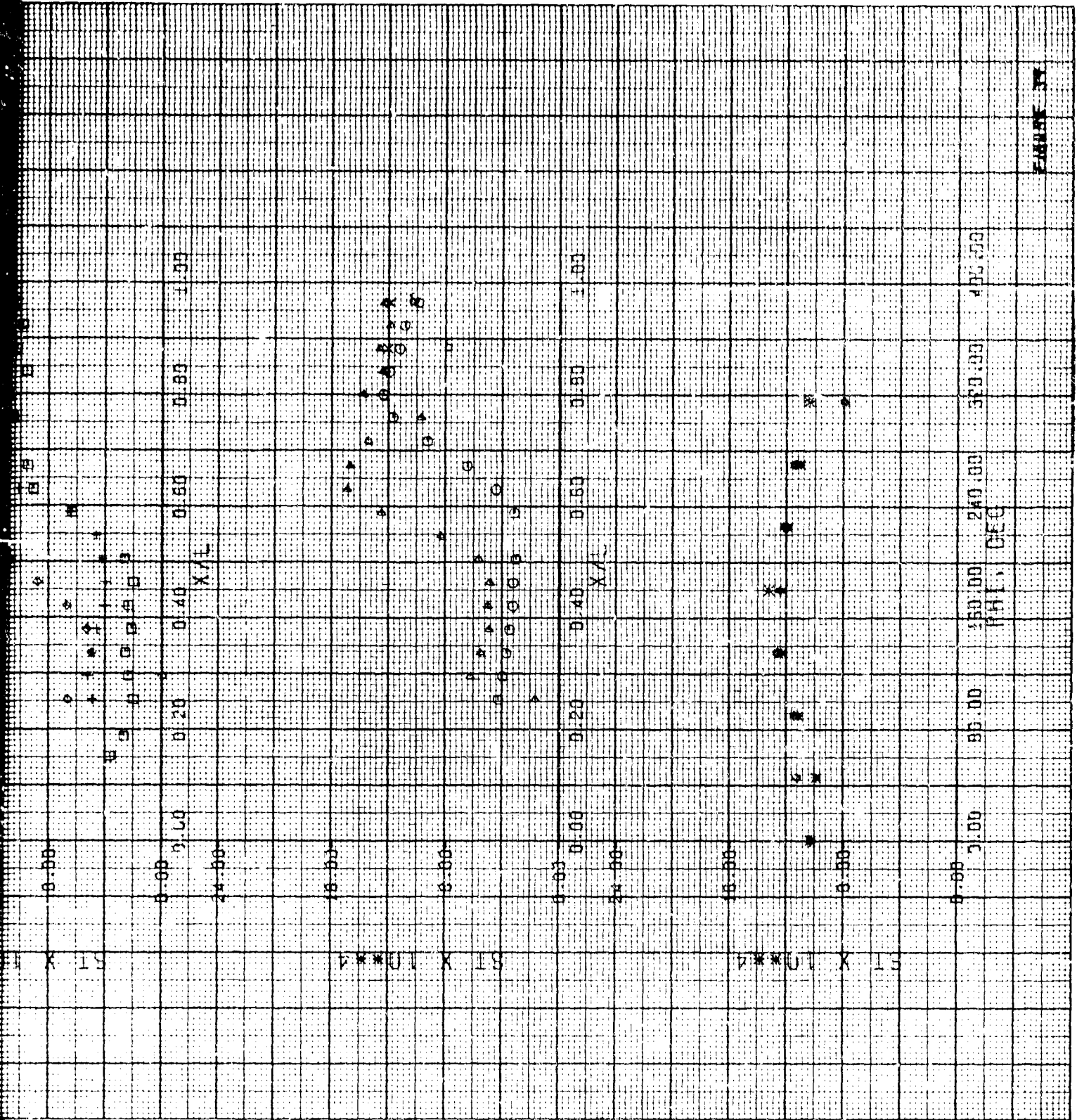


Figure 39





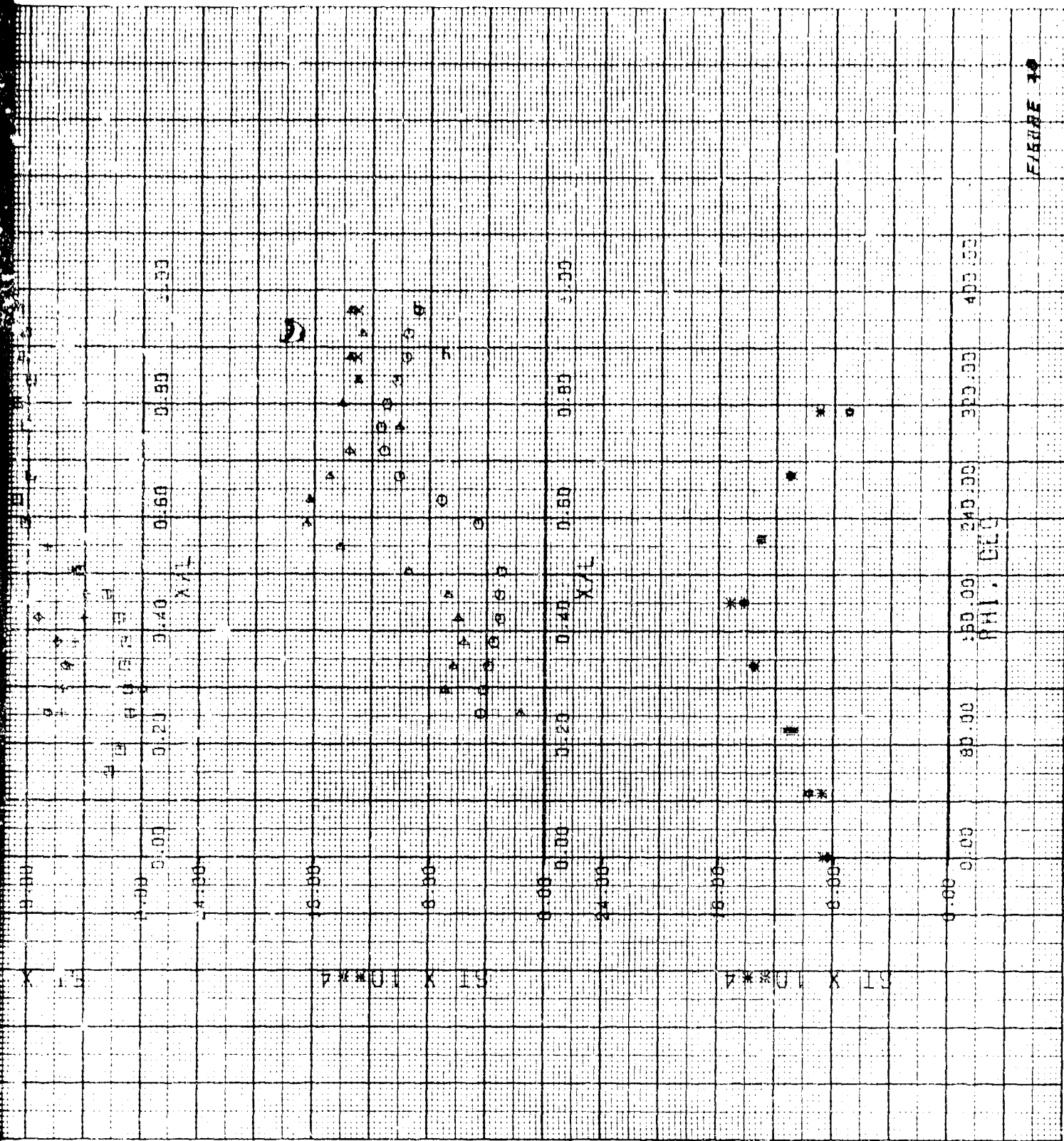


FIGURE 10

Figure 40

7/16/69

AMERICAN AIRCRAFT COMPANY  
DON HANSEN GAS DYNAMICS FACILITY  
80 INCH HYPERSONIC TUNNEL  
VT0947-000

Table with columns: ALPHA-N, BETA-N, ALPHA-T, MACH, DATE, MACH, PO, TO, TIME. Rows contain numerical data for various parameters across different test conditions.

ASYMMETRIC TRIM TEST  
HEAT TRANSFER PHASE  
SYM PHYL SYM PHL SYM

OF ASYMMETRIC TRIM TEST  
HEAT TRANSFER PHASE

GROUP 0  
MECH. NO. 13.76 X 10\*\*5  
CONFD. 3  
ALPHA 4  
TIME 0 SEC

SYN PHI 77  
SYM PHI 77  
SYN PHI 225  
SYM PHI 225  
SYN PHI 45  
SYM PHI 45  
SYN PHI 90  
SYM PHI 90  
SYN PHI 135  
SYM PHI 135  
SYN PHI 180  
SYM PHI 180

C-88  
D-88

12-88

14-88

15-88

18-88

20-88

21-88

16-88

2

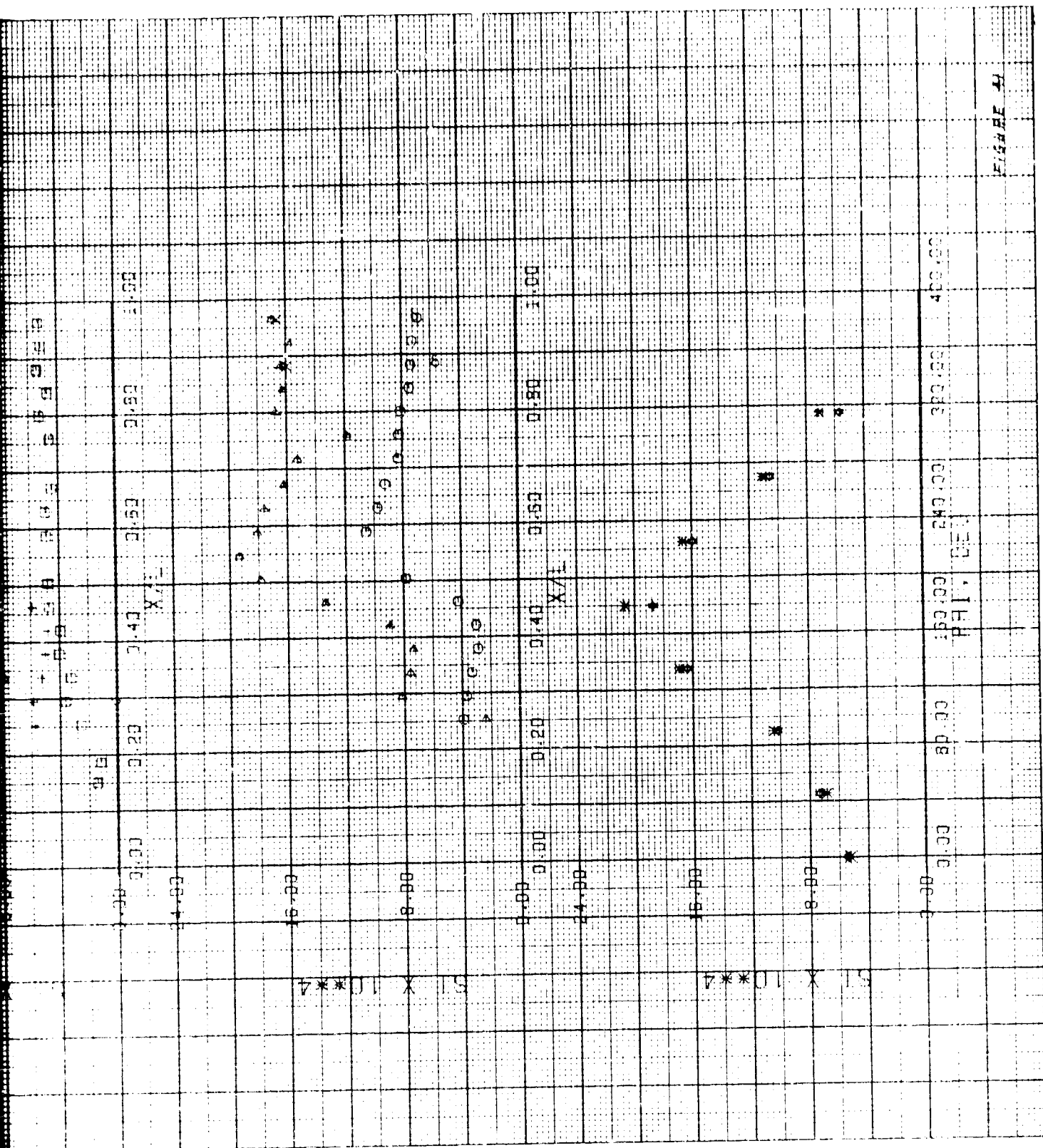


FIGURE 4

Figure 41

33

ADC-CARD INC. 21 AMMUNITION TENNESSEE  
VON HARMAN GAS DYNAMICS FACILITY  
NO INCH HYPERSONIC TUNNEL H  
HYPERSONIC

Table with columns: GROUP, COMP ID, TO, ALPHA-H, BETA-H, ALPHA-A, HULL, ALPHA-P, DATE, MACH NO, P, TO, TIME. Contains 100 rows of data points for various test conditions.

Vertical table with columns: GROUP, COMP ID, TO, ALPHA-H, BETA-H, ALPHA-A, HULL, ALPHA-P, DATE, MACH NO, P, TO, TIME. Similar structure to the main table, oriented vertically.



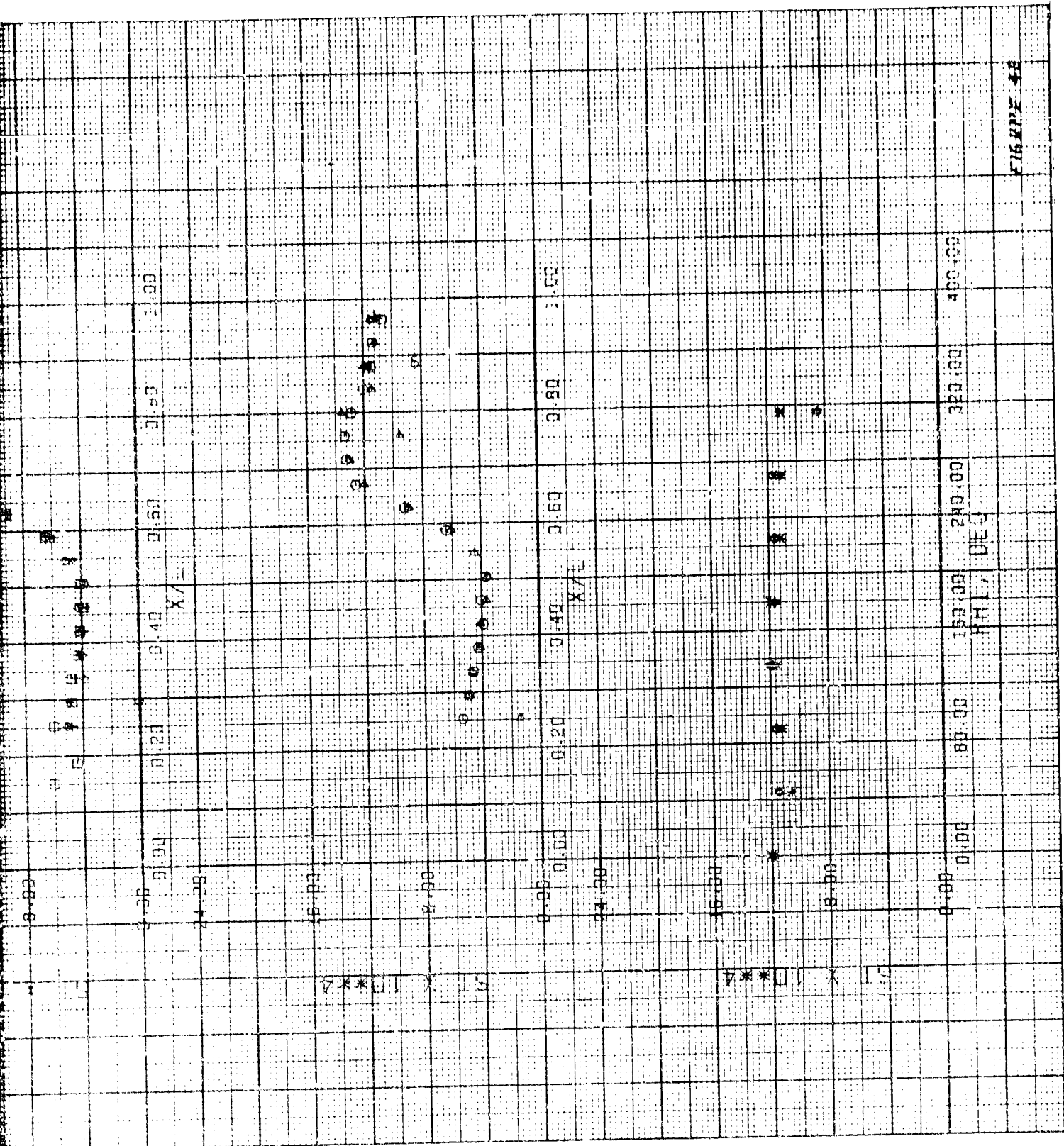


FIGURE 42

Figure 42

12/20/67

MECHANICAL ANALYSIS TENSILE  
DOW HANNA GAS DYNAMICS FACILITY  
RE ENCL HYPERBOLIC TENSILE  
VTUBAL-HUM

Table with columns: ALPH-A, ALPH-B, ALPH-C, ALPH-D, ALPH-E, ALPH-F, ALPH-G, ALPH-H, ALPH-I, ALPH-J, ALPH-K, ALPH-L, ALPH-M, ALPH-N, ALPH-O, ALPH-P, ALPH-Q, ALPH-R, ALPH-S, ALPH-T, ALPH-U, ALPH-V, ALPH-W, ALPH-X, ALPH-Y, ALPH-Z. Includes numerical data for various parameters and a vertical label on the right: SE ASYMMETRIC TUBAL TEST - HEB TRANSFER PHASE

SE ASYMMETRIC TRIM TEST

HEAT TRANSFER PHASE

GROUP 17  
 TACH NO 8  
 RE NO = 3.05 X 10\*\*5  
 CONFIG B  
 ALPHA = 0  
 TIME 0 SEC

SYM	PHI	X/L	SYM	PHI	X/L
H	0	-	X	225	-
0	45	-	H	225	-
4	90	-	0	225	-
0	135	-	0	225	-
0	180	-	0	225	-

00:00

01:00

02:00

03:00

04:00

05:00

06:00

10\*\*4

10\*\*4

2

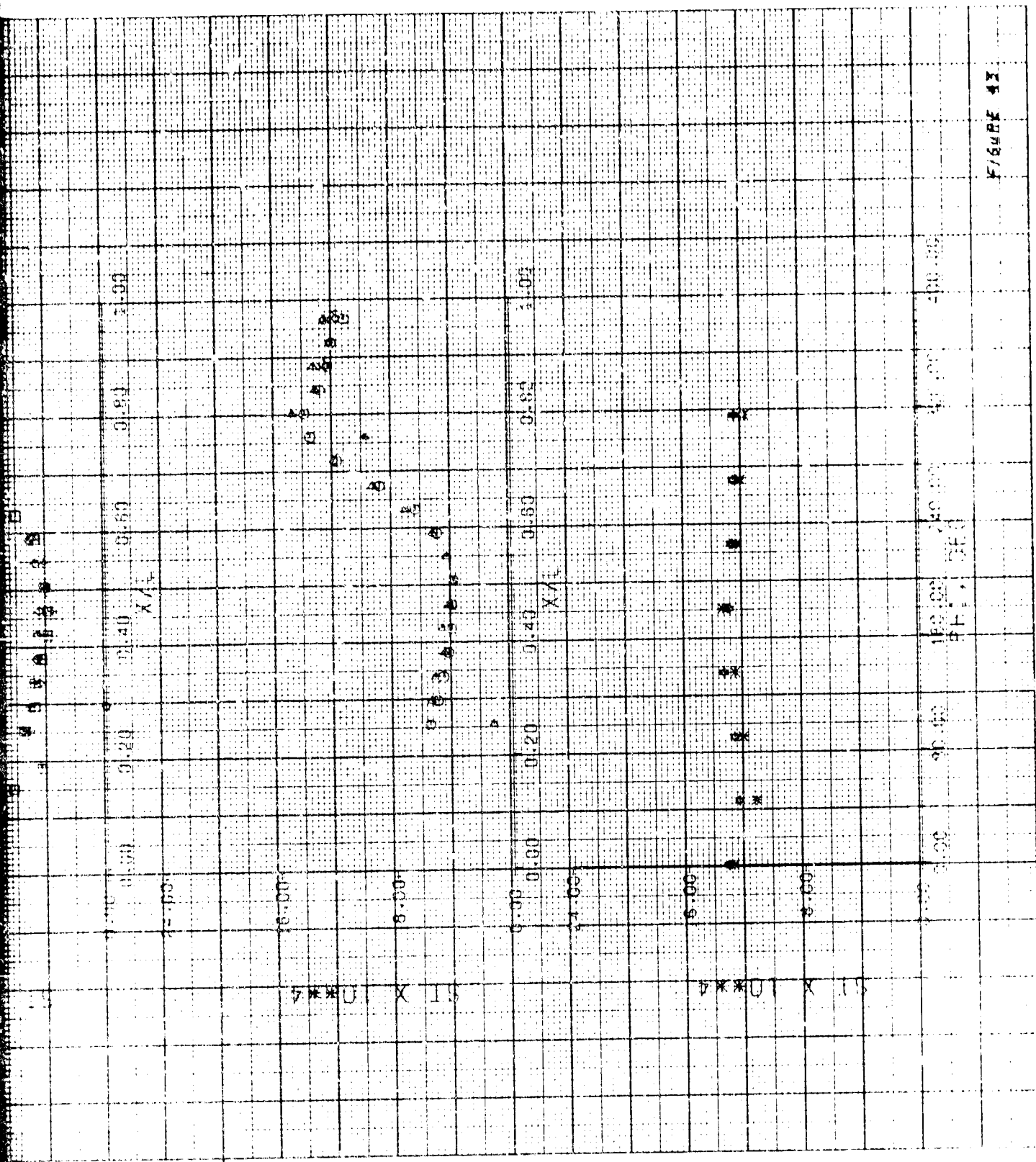


FIGURE 43

Figure 43

7/15/99

ADMINISTRATIVE SHEET - AMMOLO APS, TENNESSEE  
WERNER HARMAN GAS DYNAMICS FACILITY  
NO. INCH HYPERSONIC TUNNEL  
W70967-00

Table with columns: POINT, ALPHA-X, BETA-X, ALPHA-Z, RHO-X, RHO-Y, RHO-Z, DATE, CORR NO, PH, TO, TIME. It contains a grid of numerical data points for various coordinates and parameters.

HEAT TRANSFER PHASE  
IRIN TEST  
ASYMMETRIC IRIN TEST  
PHI XYL SYM PH YXL SYM PH XYL SYM PH

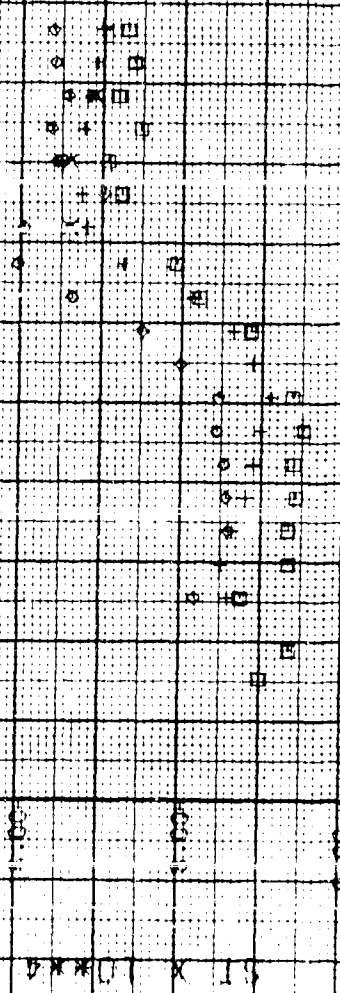
GROUP 12

AS YMMETRII TRIM TEST  
HEAT TRANSFER PHASE

GROUP 18  
MCCF NO - 8  
RE NO - 3-07 X 10\*\*6  
CONDIG 3  
SURFA 1  
TIME 0 SEC

SYM PHI XAL  
X 225  
X 270  
O 315  
O -  
X -  
X 0.88  
X 0.96

SYM PHI XAL  
O 0  
O 45  
O 90  
O 135  
O 180



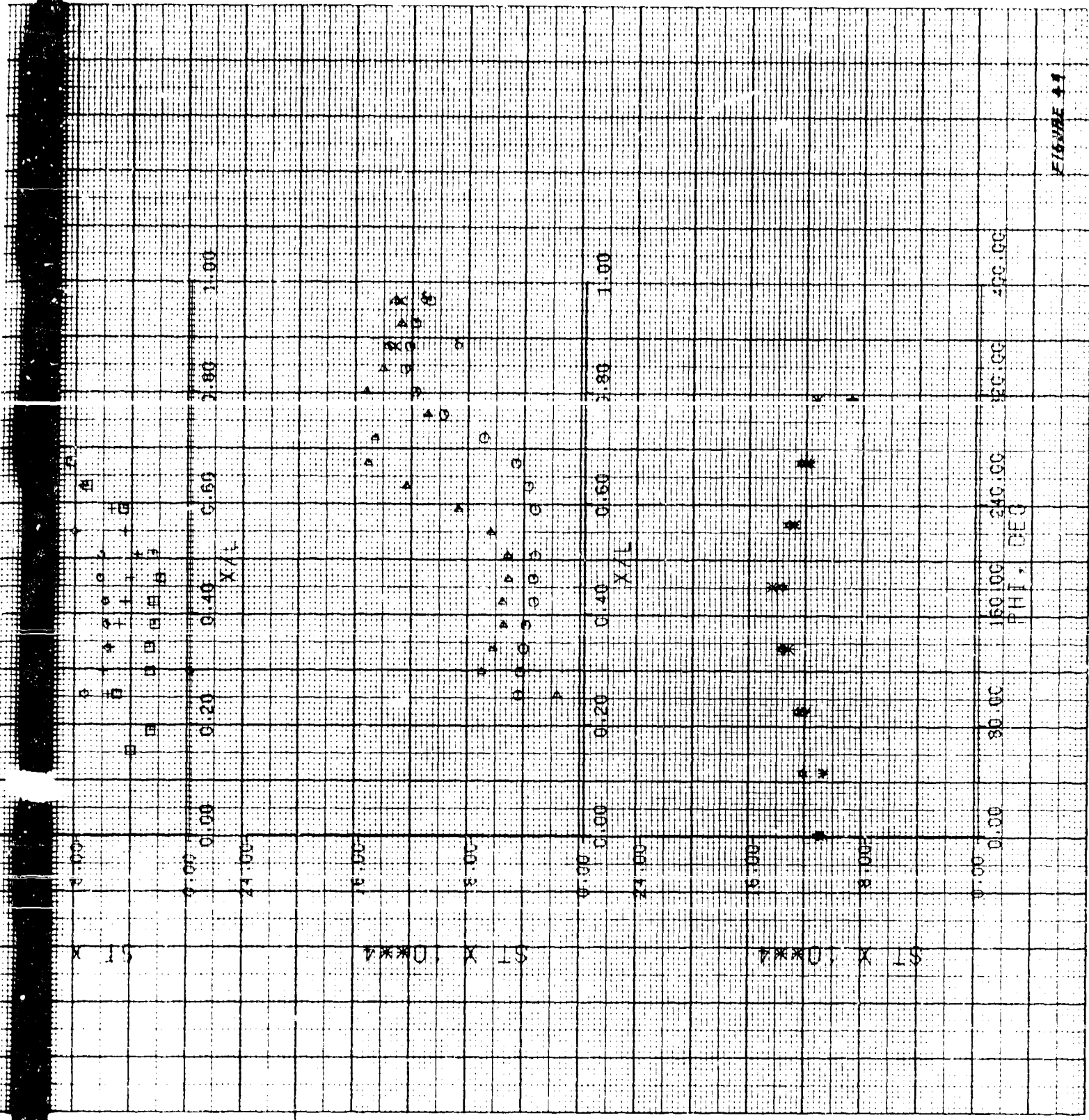


FIGURE 44

Figure 44

69

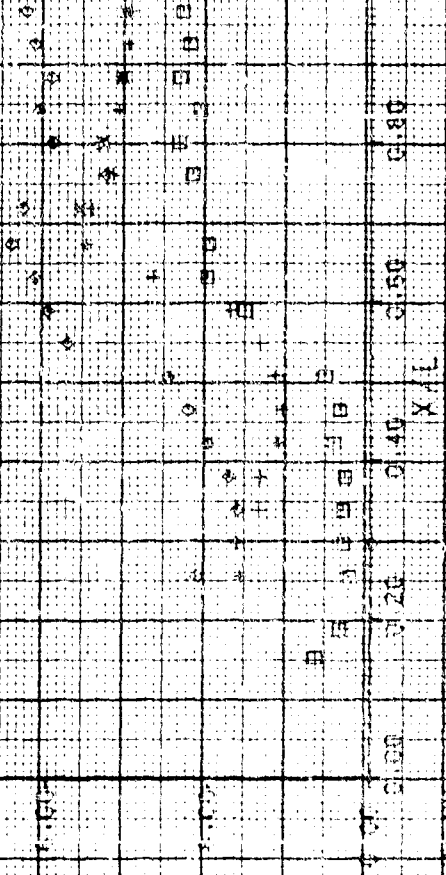


TIME  
0.00  
0.00  
0.00

GE ASYMMETRIC TRIM TEST  
HEAT TRANSFER PHASE

GROUP	NO	SYM	PHI	X/L	SYM	PHI	X/L
1	1	B	0		X	225	
2	2	B	45		X	270	
3	3	B	90		X	315	
4	4	B	135				0.80
5	5	B	180				0.90

GROUP NO: 10\*\*6  
CENT TO S  
D: 2  
TIME: 0 - SEC



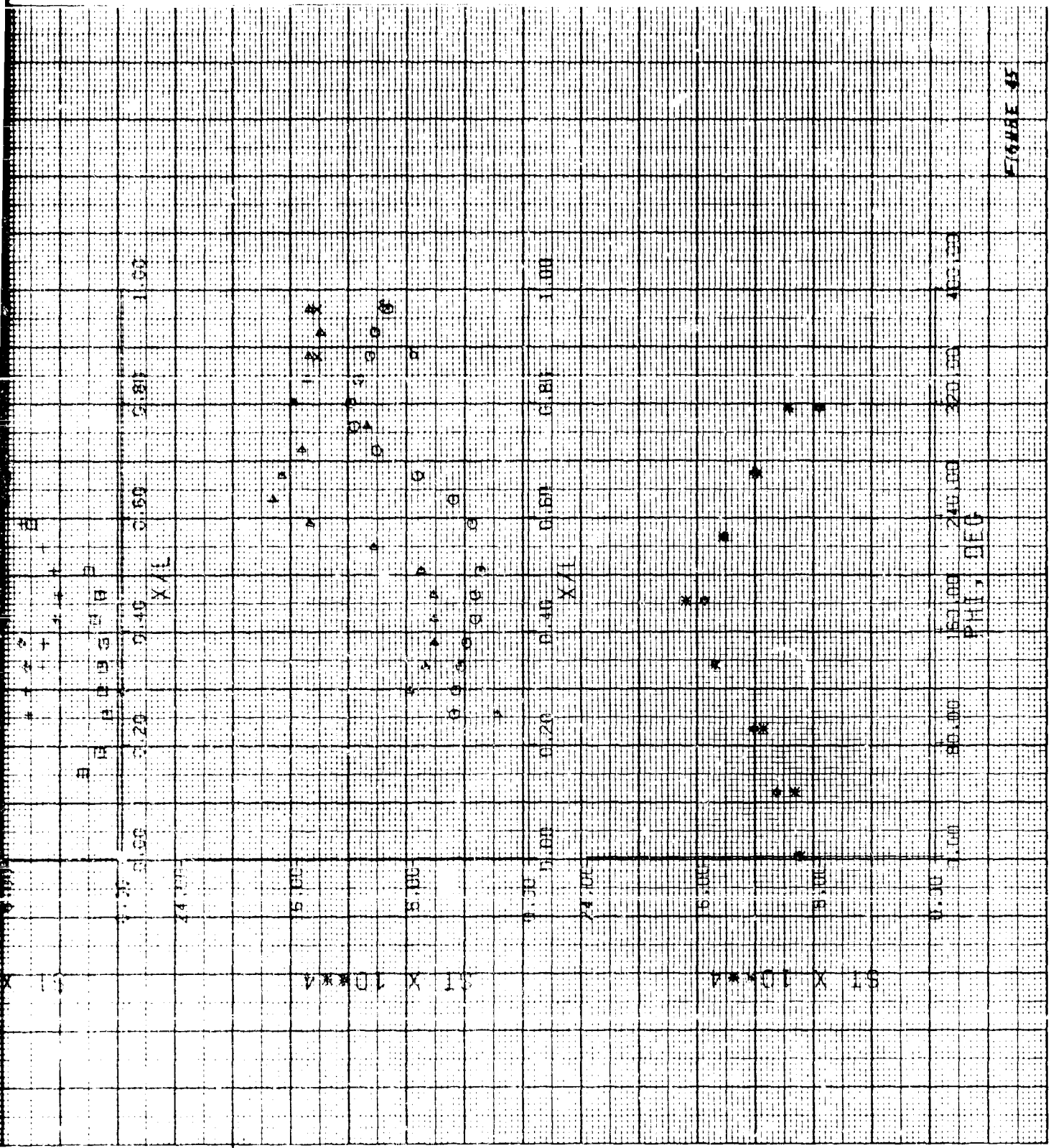


Figure 45

3

7/16/69

AEDCI AND INC - 7 ANNUL AT 11 TEMPERATURE  
WON KARMAN GAS DYNAMICS FACILITY  
30 INCH HYPERSONIC TUNNEL #  
WTOB67-000

GROUP	LINE IN	ALPHA-M	BETA-M	ALPHA-N	MU	ALPHA-P	DATE	MACH NO	PO	TO	TIME
28	3	4.02	0	4.02	0	0	716	8.00	667.H7	1313.91	00
T-IMP	P-IMP	Q-IMP	V-IMP	W-IMP	H-IMP	U-IMP					
9.9216E-01	6.8496E-02	1.0047E-00	1.8249E-01	6.0223E-05	7.6654E-08			1.0000E-00	2.7372E-06	3.2100E-02	
TC NO	X/L	PHI	TEMP	SLOPE	Q-STAG	HT	HH	HW	BT		
1	1.100	0	903.92	1.4001E-00	7.7895E-01	2.0051E-00	1.1070E-03	1.2093E-02	1.3401E-04		
2	1.007	0	491.15	4.4829E-01	1.9009E-01	2.3177E-00	9.1504E-04	1.1790E-02	1.2000E-04		
3	2.524	0	500.00	2.1160E-01	5.9654E-01	7.3249E-00	2.4547E-03	1.2001E-02	3.9822E-04		
4	2.983	0	517.21	1.5425E-01	5.0203E-01	6.3102E-00	4.3022E-03	1.2012E-02	3.4207E-04		
5	1.113	0	516.41	1.4090E-01	5.0780E-01	6.3840E-00	2.5711E-03	1.2037E-02	3.4004E-04		
6	1.781	0	519.09	1.7023E-01	5.2970E-01	6.0702E-00	2.8800E-03	1.2040E-02	3.0217E-04		
7	4.203	0	519.01	1.4525E-01	4.5110E-01	5.6793E-00	2.2877E-03	1.2045E-02	3.0837E-04		
8	4.023	0	521.59	2.1527E-01	4.4024E-01	4.1811E-00	3.2951E-03	1.2511E-02	4.4017E-04		
9	4.043	0	522.23	2.1707E-01	7.4950E-01	4.4083E-00	3.0119E-03	1.2520E-02	5.1000E-04		
10	4.043	0	520.57	2.0100E-02	2.2212E-02	2.0352E-02	1.1310E-04	1.2723E-02	1.5380E-05		
11	4.042	0	525.21	2.7020E-01	7.4400E-01	4.4397E-00	3.4013E-03	1.2599E-02	5.1200E-04		
12	4.107	0	527.14	2.4321E-01	7.1200E-01	4.0511E-00	3.0045E-03	1.2645E-02	4.9125E-04		
13	4.122	0	528.07	2.4407E-01	6.1144E-01	7.7610E-00	3.1242E-03	1.2610E-02	4.2126E-04		
14	4.022	115.0	501.00	1.1771E-01	6.7972E-01	1.1395E-03	4.5045E-03	1.2995E-02	6.1798E-04		
15	4.043	0	535.37	2.4343E-01	6.0154E-01	6.5400E-00	3.9235E-03	1.2800E-02	4.7095E-04		
16	4.043	0	537.04	2.4917E-01	7.5524E-01	9.7200E-00	3.9143E-03	1.2800E-02	5.2704E-04		
17	4.430	0	537.07	2.4094E-01	7.1022E-01	9.2370E-00	3.7174E-03	1.2800E-02	5.0100E-04		
18	4.022	0	540.74	1.1140E-01	7.4027E-01	1.0092E-03	4.0000E-03	1.2900E-02	5.0735E-04		
19	4.022	0	541.85	1.1001E-01	7.7640E-01	1.2000E-03	4.0512E-03	1.2900E-02	5.2551E-04		
20	4.043	0	539.00	1.1388E-01	6.7000E-01	1.0072E-03	4.2940E-03	1.2900E-02	5.1000E-04		
21	4.524	45.0	491.01	2.4830E-01	6.7500E-01	6.2124E-00	3.3131E-03	1.3000E-02	4.8800E-04		
22	4.043	45.0	507.07	1.4290E-01	5.0000E-01	7.2000E-00	2.4207E-03	1.2100E-02	3.9000E-04		
23	1.363	45.0	500.01	1.7545E-01	5.2700E-01	4.5510E-00	2.0404E-03	1.2200E-02	3.5591E-04		
24	1.703	45.0	509.00	1.7700E-01	5.3010E-01	4.0433E-00	2.0770E-03	1.2200E-02	3.0091E-04		
25	4.203	45.0	511.96	1.4389E-01	4.4702E-01	5.5741E-00	2.2443E-03	1.2200E-02	3.0270E-04		
26	4.023	45.0	515.31	1.4000E-01	5.0610E-01	6.3305E-00	2.5510E-03	1.2300E-02	3.0025E-04		
27	5.043	45.0	517.21	2.4205E-01	6.0931E-01	8.4011E-00	3.3045E-03	1.2400E-02	4.5022E-04		
28	4.043	115.0	544.00	4.4340E-01	1.0903E-01	1.4178E-03	5.7011E-03	1.3000E-02	7.0076E-04		
29	4.043	45.0	521.45	4.5320E-01	1.1000E-01	1.7502E-03	7.0040E-03	1.2510E-02	9.5023E-04		
30	4.022	45.0	524.77	4.7000E-01	1.0000E-01	1.9770E-03	7.0040E-03	1.2500E-02	1.0736E-03		
31	4.722	45.0	528.43	4.3990E-01	1.5482E-01	1.7085E-03	7.9247E-03	1.2675E-02	1.0683E-03		
32	4.170	45.0	531.50	4.4057E-01	1.3072E-01	1.7476E-03	7.0300E-03	1.2700E-02	9.0029E-04		
33	4.043	45.0	533.00	4.4030E-01	1.3401E-01	1.7176E-03	6.9115E-03	1.2800E-02	9.3191E-04		
34	4.043	45.0	535.20	4.4179E-01	1.2957E-01	1.6630E-03	6.6900E-03	1.2836E-02	9.0270E-04		
35	4.043	45.0	537.00	4.4400E-01	1.2307E-01	1.5850E-03	6.3010E-03	1.2899E-02	8.0021E-04		
36	4.022	45.0	539.00	4.4627E-01	1.1730E-01	1.5104E-03	6.1015E-03	1.2900E-02	8.2246E-04		
37	4.043	45.0	541.22	4.4797E-01	1.1400E-01	1.4013E-03	5.9001E-03	1.2979E-02	8.0039E-04		
38	4.043	45.0	540.41	4.4710E-01	1.0800E-01	1.4000E-03	5.0510E-03	1.2900E-02	7.0170E-04		
39	4.524	90.0	485.21	1.6090E-01	9.9351E-01	1.1900E-03	4.0300E-03	1.1600E-02	6.5219E-04		
40	4.043	90.0	488.20	2.0000E-01	9.5000E-01	1.1770E-03	4.2900E-03	1.1900E-02	6.1000E-04		
41	4.043	90.0	500.00	2.4131E-01	8.8700E-01	8.0000E-03	4.0000E-03	1.2000E-02	5.3000E-04		
42	4.043	90.0	503.62	2.4372E-01	8.0000E-01	9.0000E-03	3.9000E-03	1.2000E-02	5.3700E-04		
43	4.043	90.0	506.24	2.4132E-01	7.4200E-01	9.1000E-03	3.7000E-03	1.2100E-02	4.9000E-04		
44	4.043	90.0	508.10	2.1000E-01	6.0700E-01	8.2000E-03	3.3300E-03	1.2192E-02	4.5015E-04		
45	4.043	90.0	511.00	1.0078E-01	6.2070E-01	1.0333E-03	4.1670E-03	1.2200E-02	5.0127E-04		
46	4.043	90.0	514.57	4.1105E-01	1.2332E-01	1.5067E-03	4.2314E-03	1.2393E-02	4.3996E-04		
47	4.043	90.0	518.90	4.1903E-01	1.7930E-01	2.2555E-03	9.0000E-03	1.2400E-02	1.2200E-03		
48	4.043	90.0	522.15	4.2700E-01	2.1150E-01	2.0712E-03	1.0700E-03	1.2526E-02	1.4502E-03		
49	4.722	90.0	527.00	4.4007E-01	2.1000E-01	2.0771E-03	1.0779E-03	1.2664E-02	1.4524E-03		
50	4.170	90.0	531.97	4.4700E-01	1.4070E-01	2.3110E-03	9.1000E-03	1.2799E-02	1.2505E-03		
51	4.043	90.0	530.37	4.6910E-01	1.6852E-01	2.1507E-03	6.0500E-03	1.2721E-02	1.1071E-03		
52	4.043	90.0	532.00	4.6000E-01	1.7413E-01	2.2242E-03	6.9730E-03	1.2779E-02	1.2096E-03		
53	4.043	90.0	534.70	4.4420E-01	1.6380E-01	2.1036E-03	6.4000E-03	1.2826E-02	1.1012E-03		
54	4.022	90.0	537.50	4.4500E-01	1.6345E-01	2.1054E-03	6.4700E-03	1.2893E-02	1.1021E-03		
55	4.215	90.0	538.07	6.7102E-01	1.6137E-01	2.0020E-03	6.3700E-03	1.2923E-02	1.1293E-03		
56	4.043	90.0	537.00	6.4000E-01	1.5839E-01	2.0410E-03	6.2130E-03	1.2800E-02	1.1071E-03		
57	4.524	135.0	470.00	1.2102E-01	3.5952E-01	4.2907E-00	1.7300E-03	1.1430E-02	2.3353E-04		
58	4.043	135.0	480.30	4.3000E-01	1.5931E-01	1.6034E-03	6.1930E-03	1.1670E-02	9.1570E-04		
59	4.043	135.0	489.90	1.4000E-01	1.2270E-01	1.4901E-03	6.0170E-03	1.1701E-02	8.1000E-04		
60	4.043	135.0	492.60	1.7000E-01	1.1777E-01	1.4000E-03	5.7000E-03	1.1800E-02	7.0070E-04		
61	4.043	135.0	496.00	1.7850E-01	1.2250E-01	1.5000E-03	6.0519E-03	1.1900E-02	6.1570E-04		
62	4.043	135.0	501.43	4.4000E-01	1.3511E-01	1.6629E-03	6.7051E-03	1.2030E-02	9.0301E-04		
63	4.043	135.0	507.23	1.5050E-01	1.9170E-01	2.3771E-00	9.5000E-03	1.2171E-02	1.2916E-03		
64	4.043	135.0	512.07	1.0000E-01	2.0510E-01	3.3072E-03	1.3370E-03	1.2206E-02	1.7965E-03		
65	4.043	135.0	516.70	1.0000E-01	2.0691E-01	3.5993E-03	1.4501E-03	1.2398E-02	1.9546E-03		
66	4.022	135.0	518.70	1.0000E-01	2.0742E-01	3.6149E-03	1.4302E-03	1.2400E-02	1.9029E-03		
67	4.022	135.0	521.20	1.0000E-01	2.0100E-01	3.2000E-03	1.0000E-03	1.2000E-02	1.0000E-03		
68	4.170	135.0	529.00	1.4205E-01	2.4032E-01	3.0049E-03	1.2339E-03	1.2707E-02	1.6033E-03		
69	4.043	135.0	524.31	4.5245E-01	1.8300E-01	2.3105E-03	9.3300E-03	1.2577E-02	1.2500E-03		
70	4.043	135.0	530.50	4.1170E-01	2.5621E-01	3.2070E-03	1.3147E-03	1.2726E-02	1.7700E-03		
71	4.043	135.0	532.62	4.4000E-01	2.3707E-01	3.0344E-03	1.2210E-03	1.2775E-02	1.6005E-03		
72	4.022	135.0	534.67	4.0000E-01	2.4250E-01	3.1132E-03	1.2570E-03	1.2823E-02	1.6090E-03		
73	4.215	135.0	536.50	4.4455E-01	2.3186E-01	2.9827E-03	1.2004E-03	1.2808E-02	1.6100E-03		
74	4.043	135.0	539.07	1.0365E-01	2.4925E-01	3.2202E-03	1.2947E-03	1.2947E-02	1.7466E-03		
75	4.524	180.0	477.29	5.4310E-01	1.0071E-01	1.9209E-03	7.7507E-03	1.1402E-02	1.0454E-03		
76	4.043	180.0	491.95	4.1927E-01	1.9855E-01	1.3070E-03	5.7907E-03	1.1000E-02	7.0057E-04		
77	4.043	180.0	491.00	4.1000E-01	1.2970E-01	1.5700E-03	6.3000E-03	1.1000E-02	6.5000E-04		
78	4.043	180.0	491.50	4.1000E-01	1.3000E-01	1.5700E-03	6.3000E-03	1.1000E-02	6.5000E-04		
79	4.043	180.0	501.00	4.3000E-01	2.4900E-01	3.0000E-03	1.2000E-03	1.2000E-02	1.2000E-03		
80	4.043	180.0	507.27	4.7300E-01	2.4900E-01	3.0000E-03	1.2000E-03	1.2000E-02	1.2000E-03		
81	4.043	180.0	509.00	4.1000E-01	2.4820E-01	3.0000E-03	1.2000E-03	1.2000E-02	1.2000E-03		
82	4.043	180.0	518.11	1.7775E-01	3.3920E-01	4.2000E-03	1.7173E-03	1.2400E-02	2.3100E-03		
83	4.043	180.0	520.30	1.6932E-01	2.2030E-01	3.0000E-03	1.0700E-03	1.2400E-02	1.9000E-03		
84	4.102	180.0	524.07	1.5000E-01	2.8100						

GROUP 20  
MACH NO 8

SYM PHI X/Y  
SYM PHI X/Y

SYM PHI X/Y  
SYM PHI X/Y

SYM PHI X/Y  
SYM PHI X/Y

SYM PHI X/Y  
SYM PHI X/Y

ASYMMETRIC TRIM TEST  
HEAT TRANSFER PHASE

GROUP	20	SYM	PHI	X/Y	SYM	PHI	X/Y
MACH NO	8		0	-	*	225	-
REF NO	3.01 X 10*6		45	-	*	270	-
CENTRO	3		90	-	*	315	-
ALPHA	4		135	-	*	-	0.88
TIME	1 SEC		180	-	*	-	0.86

32.00

24.00

16.00

8.00

0.00

0.20

0.40

0.60

0.80

1.00

2

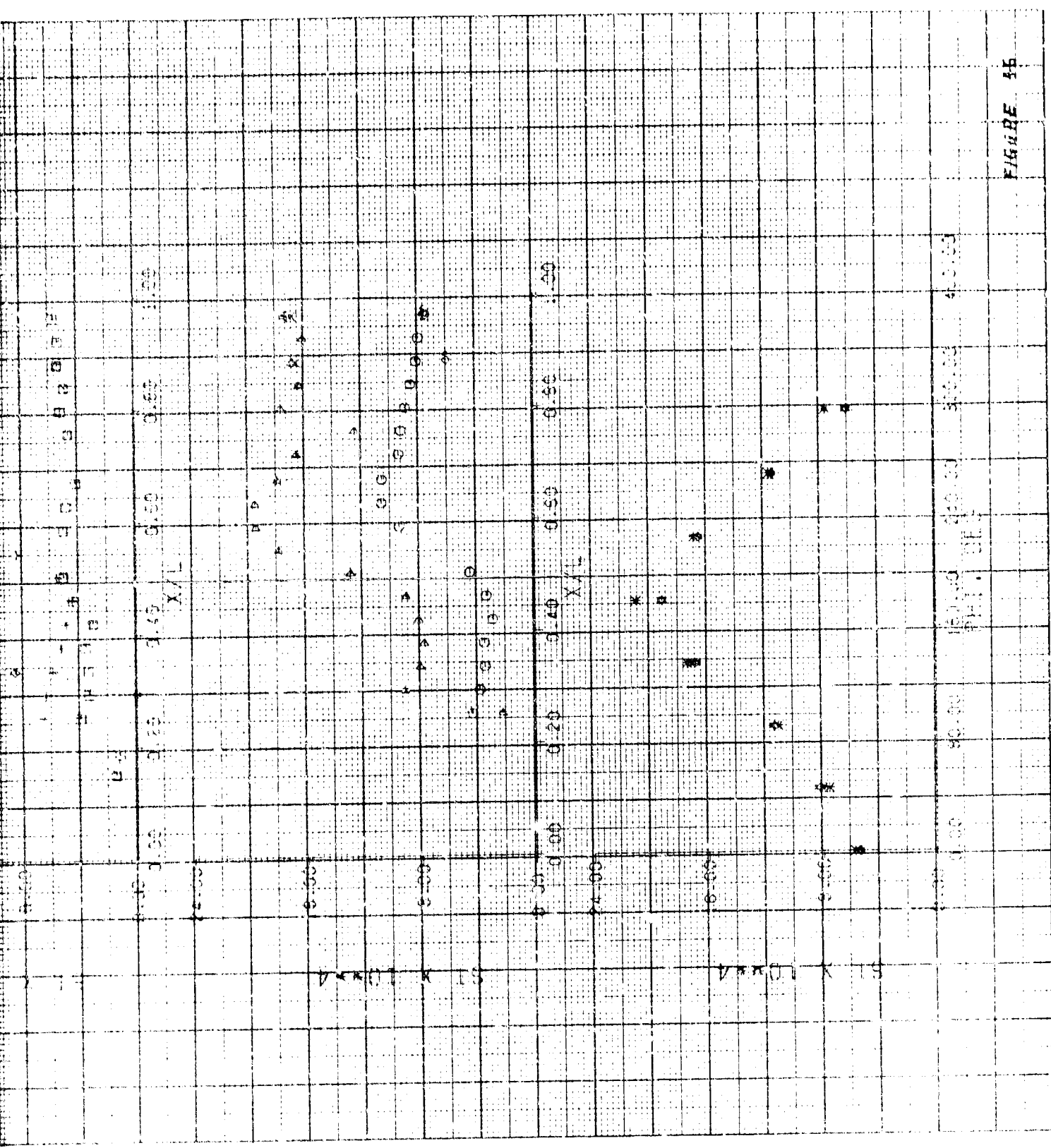


FIGURE 46

Figure 46

DEPARTMENT OF AERONAUTICS AND SPACE  
Wallops Flight Facility  
50 TON HYPERSONIC TUNNEL 3

Table with columns: GROUP, COND IN, ALPHA-N, META-N, ALPHA-S, ALL, LUMP, DATE, MAGN, PC, TO, TIME. Rows contain numerical data for various test conditions and parameters.

AS SYMMETRIC TRIM TEST  
HEB TRANSFER PHASE  
S.M. PHILIP



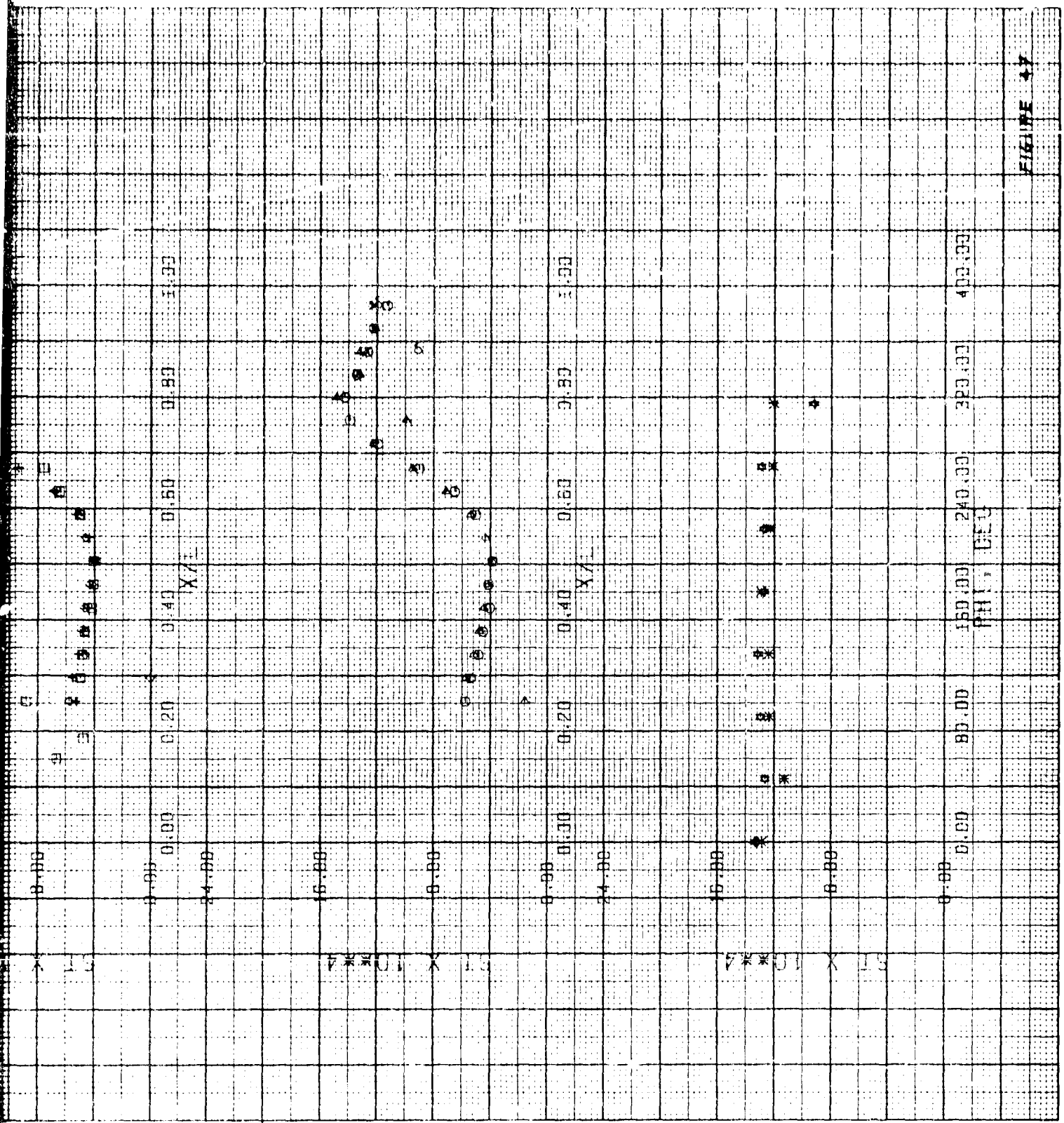


Figure 47

MEDICAL INC. 1 ANNULD AFS, TENNESSEE  
VOM KARNAN GAS DYNAMICS FACILITY  
D5 (SCH. HYPERBOLIC TUNNEL -  
V10047-000

GROUP 37 COMP 10 ALPHA-N 22 DELTA-N 8 ALPHA-T 02 RUM 0 ALP-P 0 DATE 716 MACH 40 PD 472.17 TO 1866.18 TIME 11.00  
9.34137E 01 4.07791E-02 2.17406E 00 3.70397E 03 4.30112E-00 7.52074E-00 7.20431E 00 2.02031E 00 3.14719E 02

Table with columns: GROUP, COMP, ALPHA-N, DELTA-N, ALPHA-T, RUM, ALP-P, DATE, MACH, PD, TO, TIME. Rows contain numerical data for various parameters across different stages or components.

Large grid table with multiple columns and rows, likely representing a detailed data matrix or a specific analysis table. The grid is mostly empty with some faint markings.

ASIMMETRIC TRIM TEST  
HEAT TRANSFER PHASE  
SYN PHIL XAL SYM PHE XAL  
GROUP 37  
MACH NO 18

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2  
1

GROUP	REF NO	CONFIG	ALPHA	TIME	SEC	SYM	PHI	X71	SYM	PHI	X71	TEST
27	27											
32	32											
34	34											
35	35											
38	38											
44	44											
45	45											
46	46											
47	47											
48	48											
49	49											
50	50											
51	51											
52	52											
53	53											
54	54											
55	55											
56	56											
57	57											
58	58											
59	59											
60	60											
61	61											
62	62											
63	63											
64	64											
65	65											
66	66											
67	67											
68	68											
69	69											
70	70											
71	71											
72	72											
73	73											
74	74											
75	75											
76	76											
77	77											
78	78											
79	79											
80	80											
81	81											
82	82											
83	83											
84	84											
85	85											
86	86											
87	87											
88	88											
89	89											
90	90											
91	91											
92	92											
93	93											
94	94											
95	95											
96	96											
97	97											
98	98											
99	99											
100	100											

ST X 10\*\*4

1/2

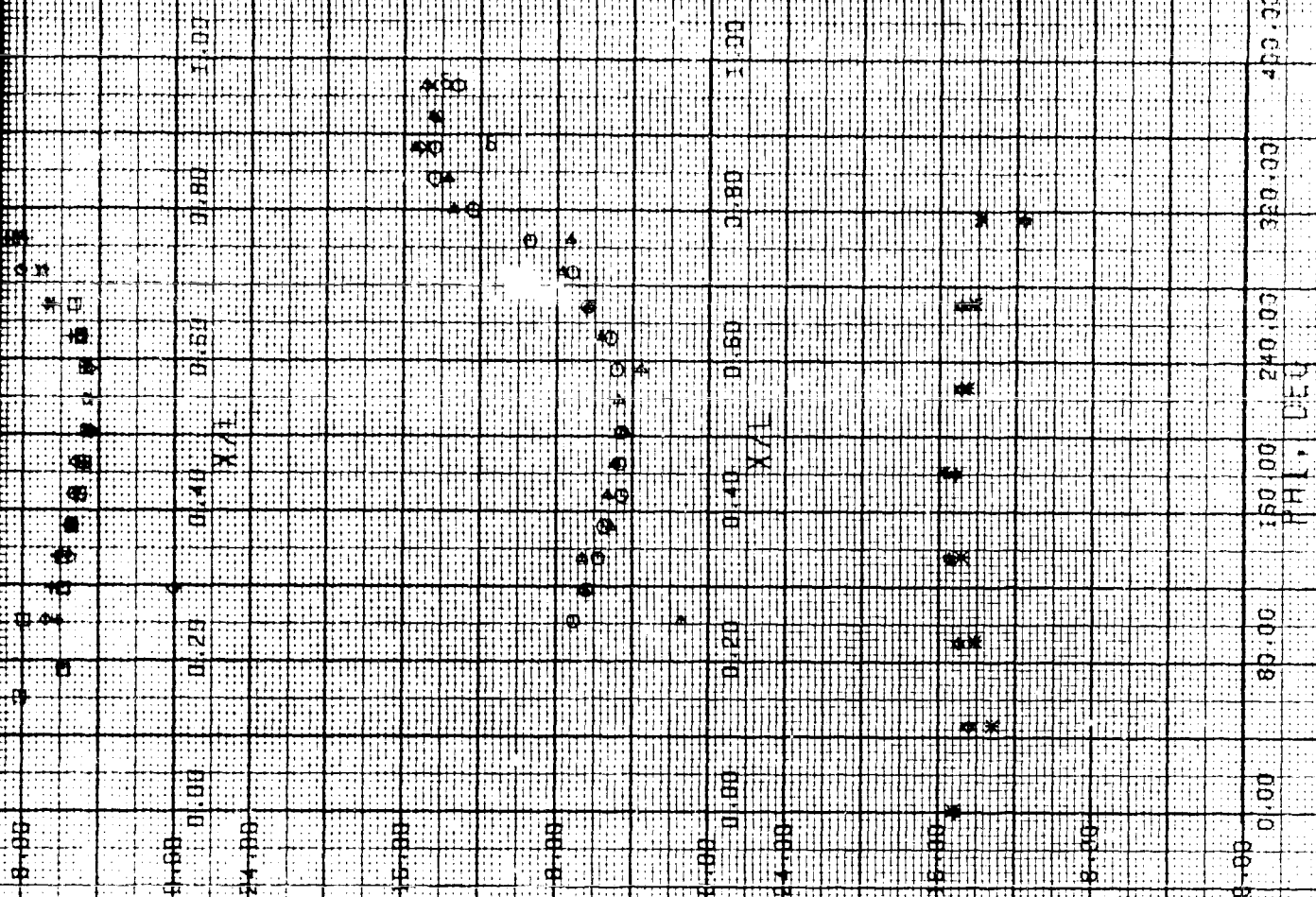


FIGURE 48

Figure 48

AEC(ARO, INC.) ANNUAL REPORT, TENNESSEE  
VON KARMAN GAS DYNAMICS FACILITY  
90 INCH HYPERSONIC TUNNEL 8  
V09947-000

Table with columns: GROUP, COND, ALPHA-M, BETA-M, ALPHA-θ, HULL, ALP-P, DATE, MACH NO, PO, TU, TIME. The table contains multiple rows of numerical data representing experimental results.

OF ASYMMETRIC TRIM TIGHT  
HEAT TRANSFER PHASE

GROUP	AS	HEAT	ASYMMETRIC	TRIM	TEST
MACH NO	TRANSFER	PHI	X/Y	SYM	PHI
28					X7E
20					225
10		45			210
9		90			315
1		135			
0		180			
0					0-90
0					0-90

TIME	PHI	X/Y	SYM	PHI	TEST
00-00					
01-00					
05-00					
09-00					
13-00					
17-00					
21-00					

2

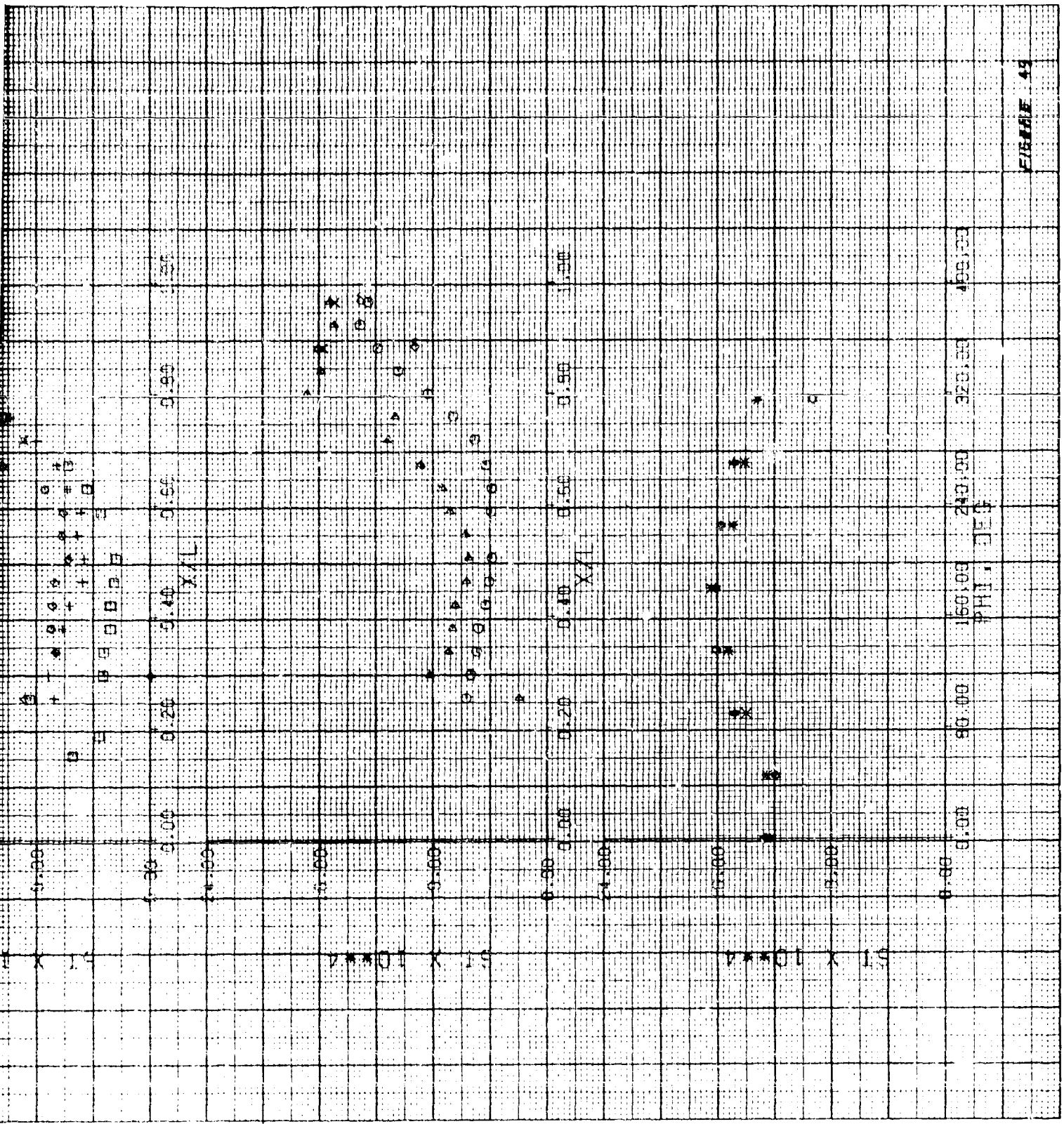


FIGURE 49

Figure 49

11

AEDC (ARO, INC.) ANNULUS AP-3 TEMPERATURE  
VON KARMAN GAS DYNAMICS FACILITY  
NO INLET HYPERSONIC TUNNEL #1  
V10067-000

GROUP COMP 16 ALPHA-H BETA-H ALPHA-H HULL ALP-P DATE MACH NO PH TO TIME  
29 3 2.05 0 2.05 0 7.14 7.00 471.37 1000.00 1.10  
1-IMP P-IMP Q-IMP V-IMP W-IMP X-IMP Y-IMP Z-IMP  
9.3623E 01 4.86757E-02 2.1722E 00 3.70021E-03 4.36264E-03 7.33749E-00 2.49227E 04 2.01073E 04 3.5390E 02

Table with columns: TC NO, R/L, PHI, TEMP, SLOPE, S-510a, HT, PH, AU, ST. Rows 1-98 containing numerical data for temperature and slope measurements.

GE ASYMMETRIC TRIM CASE  
HEAT TRANSFER EFFECT  
ASYM. PH. XI  
GROUP 29

GROUP 29

ASYM PHIL XYL SYM PHL XYL

DE ASYMMETRIC TRIM TEST  
HEAT TRANSFER PHASE

GROUP	ASYM	PHI	XYL	SYM	PHI	XYL
29						
MACH NO	1	8				
REF NO	12.19	X 10**6				
CENTIG	3					
ALPHA	2					
TIME	0	SFC				

12-10

14-10

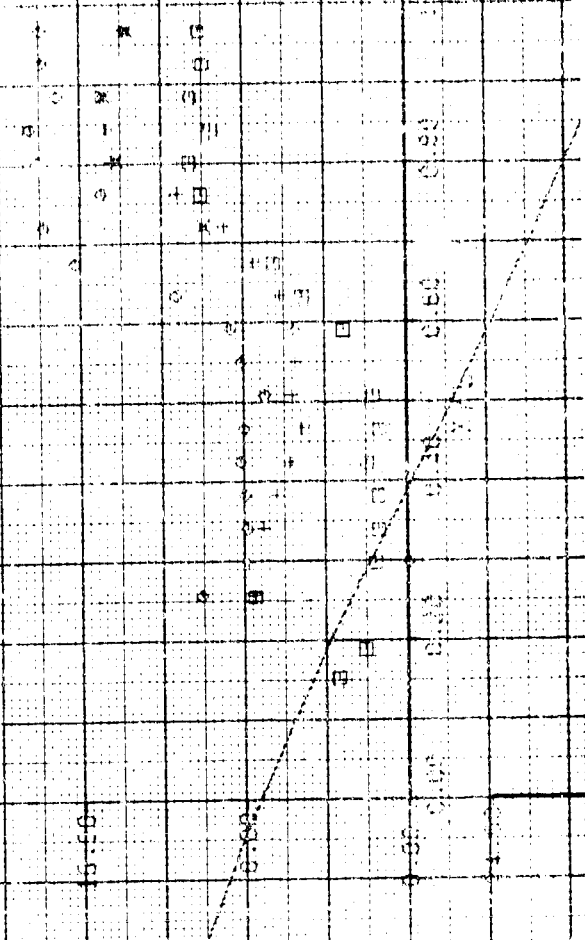
15-10

16-10

17-10

22

12\*\*04 X 10



18-10

19-10

20-10

21-10

22-10

23-10

24-10

25-10

26-10

27-10

28-10

29-10

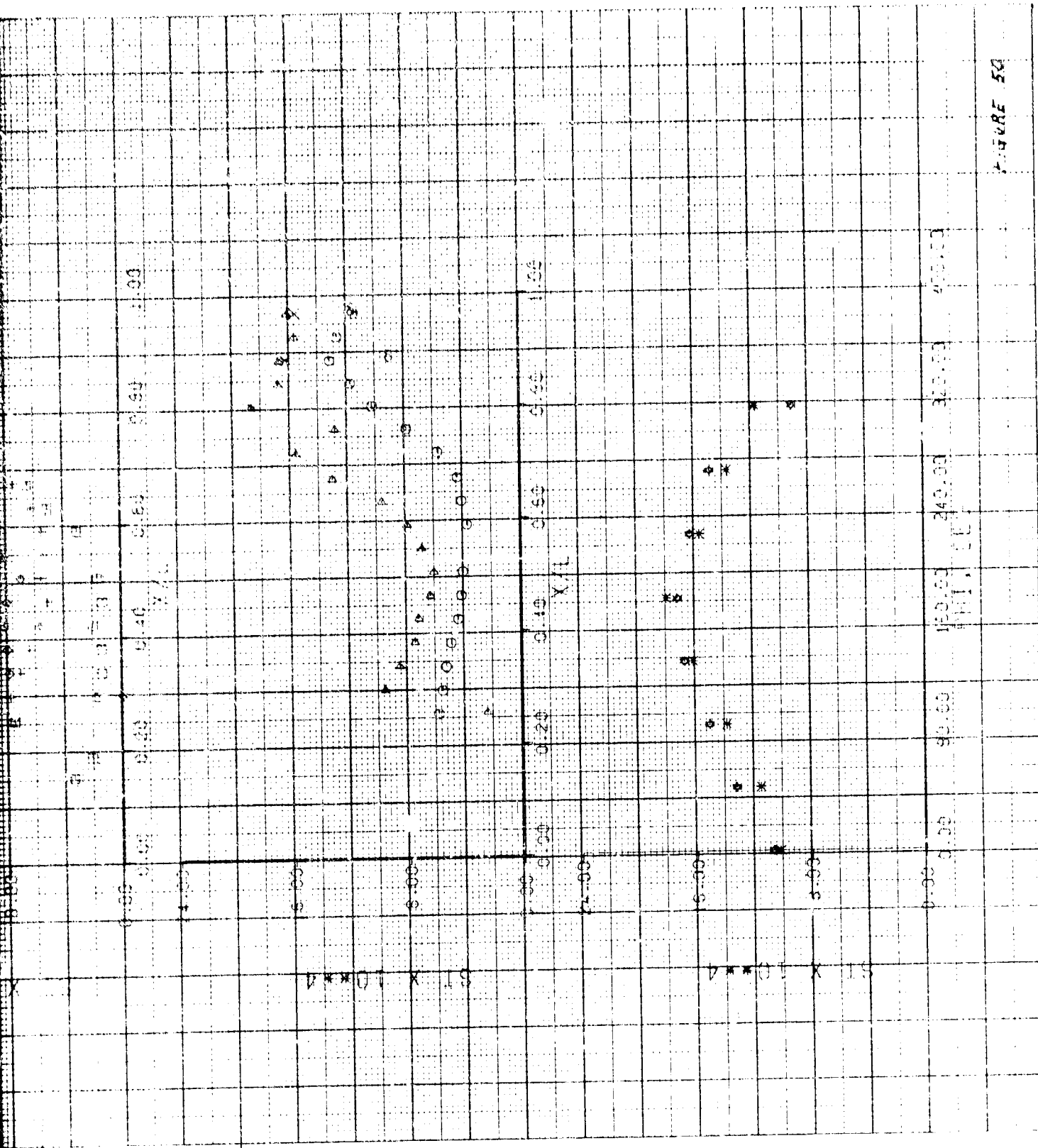


FIGURE 50

Figure 50

3

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RESEARCH AND ANALYSIS CENTER  
FOR HIGH SPEED AERODYNAMICS  
ON THE HYPERBOLIC TUNNEL

GROUP 10010 ALPHA-N ALPHA-M ALPHA-S ALPHA-T ALPHA-U ALPHA-V ALPHA-W ALPHA-X ALPHA-Y ALPHA-Z DATE MACH NO. P1 P2 TO TIME  
7.3047E-01 4.0948E-02 2.1710E-01 3.7920E-01 4.2415E-01 7.3554E-01 2.1794E-01 1.9981E-01 3.1618E-01

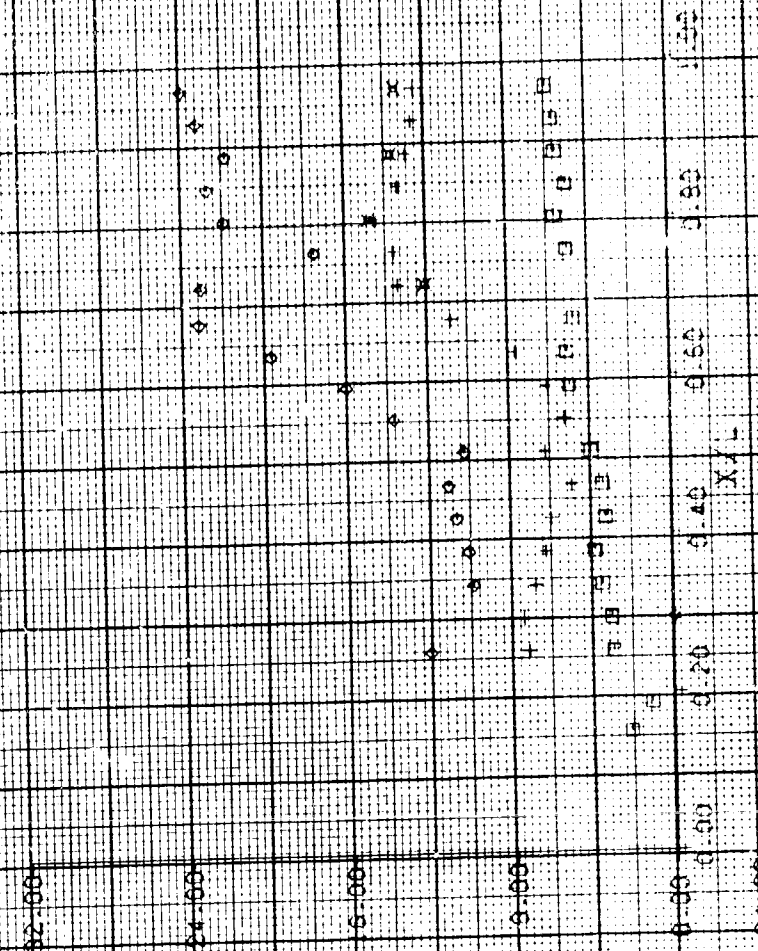
Table with columns: VC NO, RAL, PHI, TEMP, SLOPE, G-STOW, HT, HM, HW, ST. Rows contain numerical data for various test cases.

ASYNMETRIC TORM TEST  
HEAT TRANSFER PHASE  
SYM

GE ASYMMETRIC TRIM TEST  
HEAT TRANSFER PHASE

GROUP 30  
 MACH NO 8  
 RE NO 2.18 X 10\*\*6  
 CONFIG 3  
 ALPHA 4  
 TIME 0 SEC

SYM PHI XVL SYM PHI XAE  
 \* \* \* \* \*  
 \* \* \* \* \*  
 \* \* \* \* \*  
 \* \* \* \* \*  
 \* \* \* \* \*



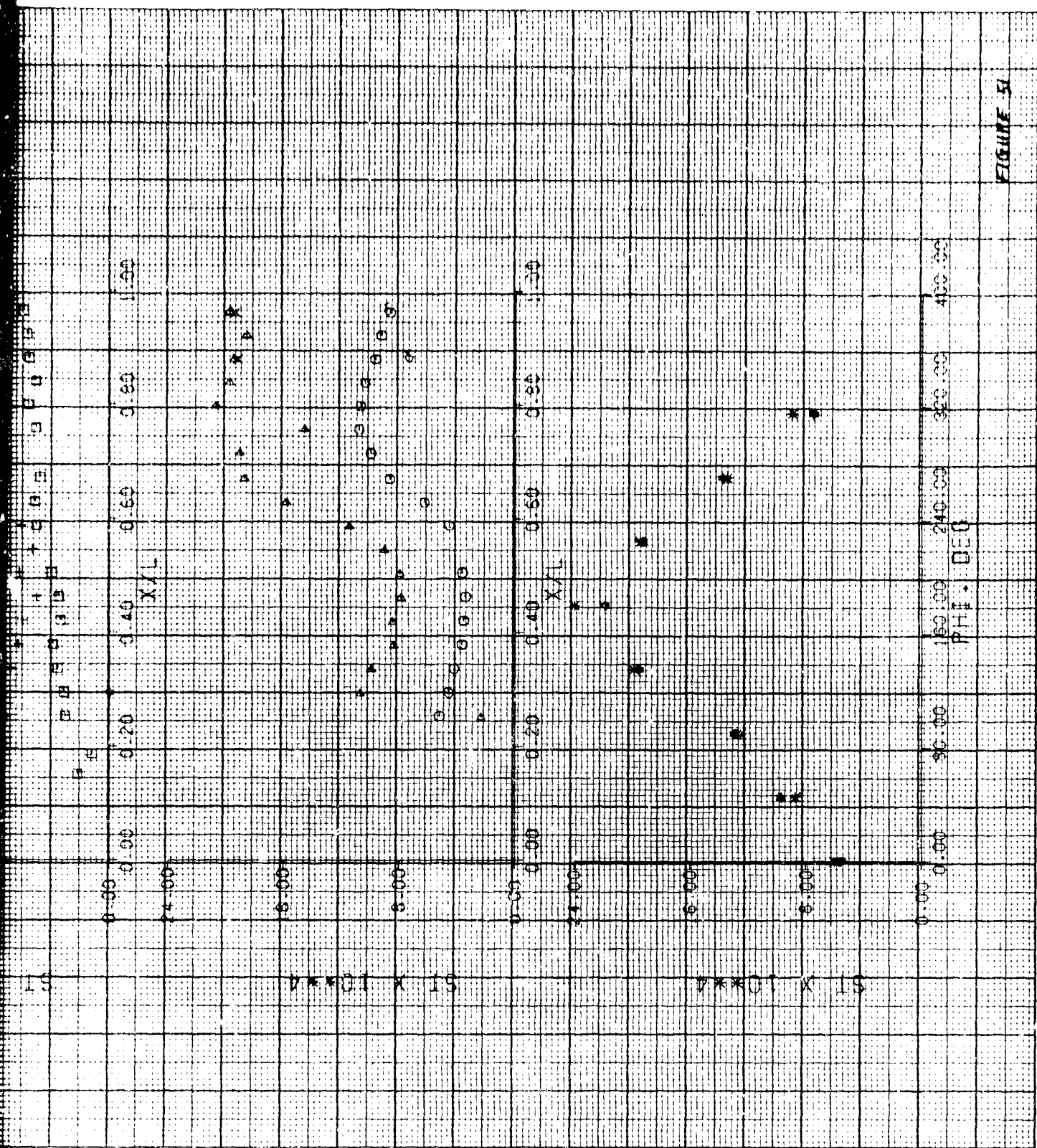


FIGURE 51

Figure 51

7/16/68

AEGICARD (INC.) ANNULAR SPIN TUNNEL  
WON KAMMAN GAS DYNAMICS FACILITY  
50 INCH HYPERSONIC TUNNEL  
WTONA7-W00

Table with columns: RUN NO, CUMUL T, ALPHA-0, BETA-0, ALPHA-0, HULL, ALPHA-0, DATE, PACH NO, PH, TO, TIME. Rows contain numerical data for various test runs.

GE ASYMMETRIC RIM FILE  
HEAT TRANSFER PHASE  
SYM PHASE

GF	ASYMMETRIC	TRIM	TEST
HEAT	TRANSFER	PHASE	
1500P	SYM	YM	SYM
1600P	PHI	PHI	PHI
1700P	PHI	PHI	PHI
1800P	PHI	PHI	PHI
1900P	PHI	PHI	PHI
2000P	PHI	PHI	PHI
2100P	PHI	PHI	PHI
2200P	PHI	PHI	PHI
2300P	PHI	PHI	PHI
2400P	PHI	PHI	PHI
2500P	PHI	PHI	PHI
2600P	PHI	PHI	PHI
2700P	PHI	PHI	PHI
2800P	PHI	PHI	PHI
2900P	PHI	PHI	PHI
3000P	PHI	PHI	PHI
3100P	PHI	PHI	PHI
3200P	PHI	PHI	PHI
3300P	PHI	PHI	PHI
3400P	PHI	PHI	PHI
3500P	PHI	PHI	PHI
3600P	PHI	PHI	PHI
3700P	PHI	PHI	PHI
3800P	PHI	PHI	PHI
3900P	PHI	PHI	PHI
4000P	PHI	PHI	PHI
4100P	PHI	PHI	PHI
4200P	PHI	PHI	PHI
4300P	PHI	PHI	PHI
4400P	PHI	PHI	PHI
4500P	PHI	PHI	PHI
4600P	PHI	PHI	PHI
4700P	PHI	PHI	PHI
4800P	PHI	PHI	PHI
4900P	PHI	PHI	PHI
5000P	PHI	PHI	PHI
5100P	PHI	PHI	PHI
5200P	PHI	PHI	PHI
5300P	PHI	PHI	PHI
5400P	PHI	PHI	PHI
5500P	PHI	PHI	PHI
5600P	PHI	PHI	PHI
5700P	PHI	PHI	PHI
5800P	PHI	PHI	PHI
5900P	PHI	PHI	PHI
6000P	PHI	PHI	PHI
6100P	PHI	PHI	PHI
6200P	PHI	PHI	PHI
6300P	PHI	PHI	PHI
6400P	PHI	PHI	PHI
6500P	PHI	PHI	PHI
6600P	PHI	PHI	PHI
6700P	PHI	PHI	PHI
6800P	PHI	PHI	PHI
6900P	PHI	PHI	PHI
7000P	PHI	PHI	PHI
7100P	PHI	PHI	PHI
7200P	PHI	PHI	PHI
7300P	PHI	PHI	PHI
7400P	PHI	PHI	PHI
7500P	PHI	PHI	PHI
7600P	PHI	PHI	PHI
7700P	PHI	PHI	PHI
7800P	PHI	PHI	PHI
7900P	PHI	PHI	PHI
8000P	PHI	PHI	PHI
8100P	PHI	PHI	PHI
8200P	PHI	PHI	PHI
8300P	PHI	PHI	PHI
8400P	PHI	PHI	PHI
8500P	PHI	PHI	PHI
8600P	PHI	PHI	PHI
8700P	PHI	PHI	PHI
8800P	PHI	PHI	PHI
8900P	PHI	PHI	PHI
9000P	PHI	PHI	PHI
9100P	PHI	PHI	PHI
9200P	PHI	PHI	PHI
9300P	PHI	PHI	PHI
9400P	PHI	PHI	PHI
9500P	PHI	PHI	PHI
9600P	PHI	PHI	PHI
9700P	PHI	PHI	PHI
9800P	PHI	PHI	PHI
9900P	PHI	PHI	PHI
10000P	PHI	PHI	PHI

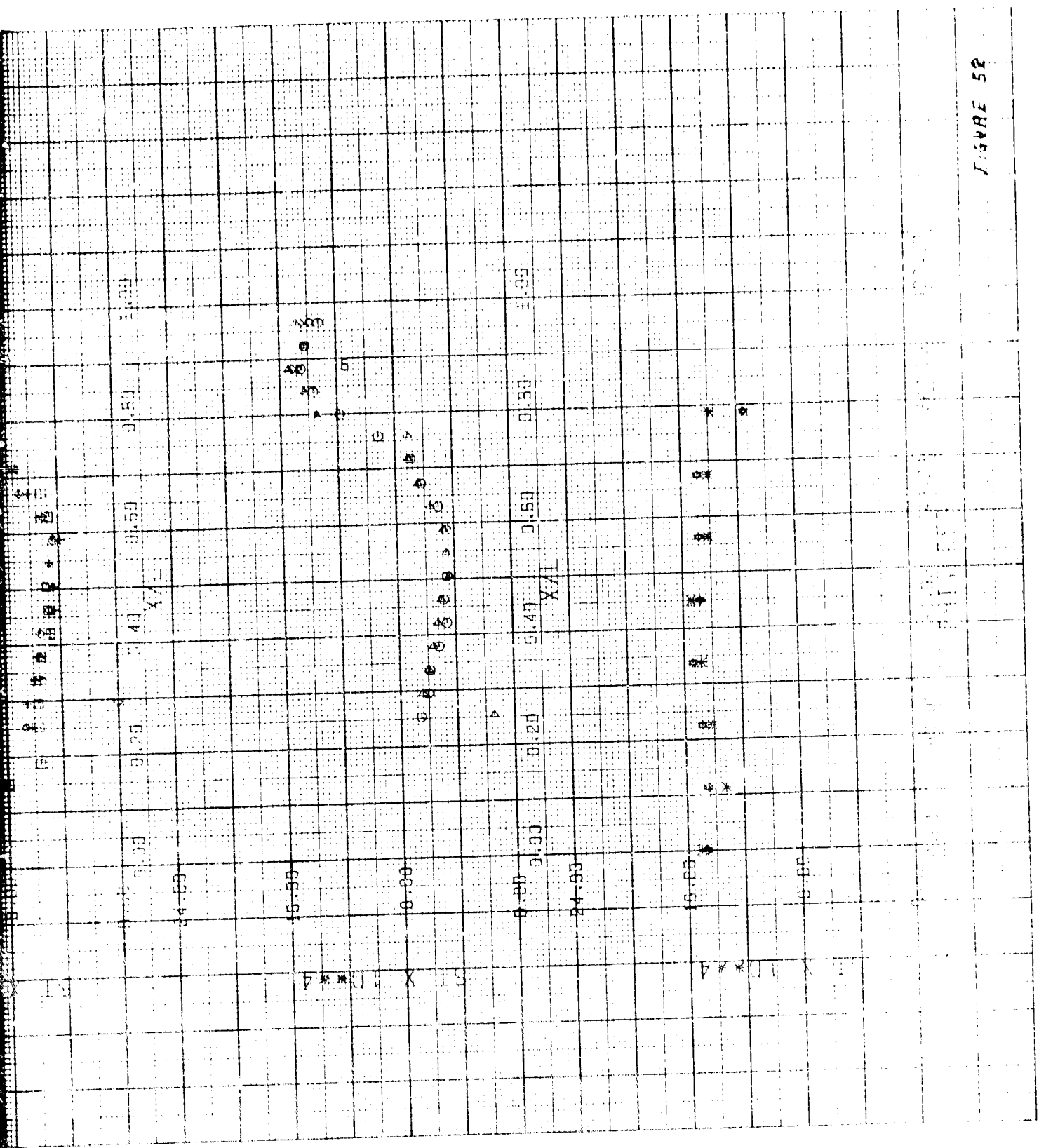


FIGURE 52

Figure 52

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AEC (AMU) INC. - ARMOED AF 50 TENNESSEE  
WIND TUNNEL GAS DYNAMICS FACILITY  
50 INCH HYPERSONIC TUNNEL  
WIND TUNNEL

PHI	TEMP	SEOP	Q-570	HT	HI	ST
1	910.54	2.4280E-08	3.1081E-01	1.7827E-04	3.1307E-03	9.6976E-04
2	911.55	1.4444E-08	4.0597E-01	5.5113E-04	2.2665E-03	6.8768E-04
3	912.56	2.4444E-08	5.1257E-01	1.0216E-03	5.1171E-03	1.2731E-03
4	913.57	1.4444E-08	6.2509E-01	2.0056E-04	2.3819E-03	1.2299E-03
5	914.58	1.4444E-08	7.4514E-01	4.2361E-04	2.5194E-03	1.2299E-03
6	915.59	1.4608E-08	8.7197E-01	5.2247E-04	2.1071E-03	1.2299E-03
7	916.60	1.4608E-08	1.0050E-01	6.0019E-04	1.9366E-03	1.2299E-03
8	917.61	1.4608E-08	1.1398E-01	6.7109E-04	2.0001E-03	1.3000E-03
9	918.62	1.4608E-08	1.2741E-01	7.3580E-04	1.9738E-03	1.3000E-03
10	919.63	1.4608E-08	1.4079E-01	7.9447E-04	1.7370E-03	1.3000E-03
11	920.64	1.4608E-08	1.5412E-01	8.4717E-04	1.5917E-03	1.3000E-03
12	921.65	1.4608E-08	1.6740E-01	8.9394E-04	1.4072E-03	1.3000E-03
13	922.66	1.4608E-08	1.8063E-01	9.3474E-04	1.1942E-03	1.3000E-03
14	923.67	1.4608E-08	1.9381E-01	9.7064E-04	1.5042E-03	1.3000E-03
15	924.68	1.4608E-08	2.0694E-01	1.0017E-03	2.1019E-03	1.3000E-03
16	925.69	1.4608E-08	2.2002E-01	1.0210E-03	1.0299E-03	1.3000E-03
17	926.70	1.4608E-08	2.3305E-01	1.0412E-03	1.7619E-03	1.3000E-03
18	927.71	1.4608E-08	2.4603E-01	1.0623E-03	1.7619E-03	1.3000E-03
19	928.72	1.4608E-08	2.5896E-01	1.0843E-03	2.3361E-03	1.3000E-03
20	929.73	1.4608E-08	2.7184E-01	1.1072E-03	2.3361E-03	1.3000E-03
21	930.74	1.4608E-08	2.8467E-01	1.1310E-03	2.3361E-03	1.3000E-03
22	931.75	1.4608E-08	2.9745E-01	1.1557E-03	2.3361E-03	1.3000E-03
23	932.76	1.4608E-08	3.1018E-01	1.1813E-03	2.3361E-03	1.3000E-03
24	933.77	1.4608E-08	3.2286E-01	1.2078E-03	2.3361E-03	1.3000E-03
25	934.78	1.4608E-08	3.3549E-01	1.2351E-03	2.3361E-03	1.3000E-03
26	935.79	1.4608E-08	3.4807E-01	1.2632E-03	2.3361E-03	1.3000E-03
27	936.80	1.4608E-08	3.6060E-01	1.2921E-03	2.3361E-03	1.3000E-03
28	937.81	1.4608E-08	3.7308E-01	1.3218E-03	2.3361E-03	1.3000E-03
29	938.82	1.4608E-08	3.8551E-01	1.3522E-03	2.3361E-03	1.3000E-03
30	939.83	1.4608E-08	3.9789E-01	1.3833E-03	2.3361E-03	1.3000E-03
31	940.84	1.4608E-08	4.1022E-01	1.4151E-03	2.3361E-03	1.3000E-03
32	941.85	1.4608E-08	4.2250E-01	1.4476E-03	2.3361E-03	1.3000E-03
33	942.86	1.4608E-08	4.3473E-01	1.4808E-03	2.3361E-03	1.3000E-03
34	943.87	1.4608E-08	4.4691E-01	1.5147E-03	2.3361E-03	1.3000E-03
35	944.88	1.4608E-08	4.5904E-01	1.5493E-03	2.3361E-03	1.3000E-03
36	945.89	1.4608E-08	4.7112E-01	1.5846E-03	2.3361E-03	1.3000E-03
37	946.90	1.4608E-08	4.8315E-01	1.6206E-03	2.3361E-03	1.3000E-03
38	947.91	1.4608E-08	4.9513E-01	1.6573E-03	2.3361E-03	1.3000E-03
39	948.92	1.4608E-08	5.0706E-01	1.6947E-03	2.3361E-03	1.3000E-03
40	949.93	1.4608E-08	5.1894E-01	1.7328E-03	2.3361E-03	1.3000E-03
41	950.94	1.4608E-08	5.3077E-01	1.7716E-03	2.3361E-03	1.3000E-03
42	951.95	1.4608E-08	5.4255E-01	1.8111E-03	2.3361E-03	1.3000E-03
43	952.96	1.4608E-08	5.5428E-01	1.8513E-03	2.3361E-03	1.3000E-03
44	953.97	1.4608E-08	5.6596E-01	1.8922E-03	2.3361E-03	1.3000E-03
45	954.98	1.4608E-08	5.7759E-01	1.9338E-03	2.3361E-03	1.3000E-03
46	955.99	1.4608E-08	5.8917E-01	1.9761E-03	2.3361E-03	1.3000E-03
47	956.00	1.4608E-08	6.0070E-01	2.0191E-03	2.3361E-03	1.3000E-03
48	957.01	1.4608E-08	6.1218E-01	2.0628E-03	2.3361E-03	1.3000E-03
49	958.02	1.4608E-08	6.2361E-01	2.1072E-03	2.3361E-03	1.3000E-03
50	959.03	1.4608E-08	6.3499E-01	2.1523E-03	2.3361E-03	1.3000E-03
51	960.04	1.4608E-08	6.4632E-01	2.1981E-03	2.3361E-03	1.3000E-03
52	961.05	1.4608E-08	6.5760E-01	2.2446E-03	2.3361E-03	1.3000E-03
53	962.06	1.4608E-08	6.6883E-01	2.2918E-03	2.3361E-03	1.3000E-03
54	963.07	1.4608E-08	6.8001E-01	2.3397E-03	2.3361E-03	1.3000E-03
55	964.08	1.4608E-08	6.9114E-01	2.3883E-03	2.3361E-03	1.3000E-03
56	965.09	1.4608E-08	7.0222E-01	2.4376E-03	2.3361E-03	1.3000E-03
57	966.10	1.4608E-08	7.1325E-01	2.4876E-03	2.3361E-03	1.3000E-03
58	967.11	1.4608E-08	7.2423E-01	2.5383E-03	2.3361E-03	1.3000E-03
59	968.12	1.4608E-08	7.3516E-01	2.5897E-03	2.3361E-03	1.3000E-03
60	969.13	1.4608E-08	7.4604E-01	2.6418E-03	2.3361E-03	1.3000E-03
61	970.14	1.4608E-08	7.5687E-01	2.6946E-03	2.3361E-03	1.3000E-03
62	971.15	1.4608E-08	7.6765E-01	2.7481E-03	2.3361E-03	1.3000E-03
63	972.16	1.4608E-08	7.7838E-01	2.8023E-03	2.3361E-03	1.3000E-03
64	973.17	1.4608E-08	7.8906E-01	2.8572E-03	2.3361E-03	1.3000E-03
65	974.18	1.4608E-08	7.9969E-01	2.9128E-03	2.3361E-03	1.3000E-03
66	975.19	1.4608E-08	8.1027E-01	2.9691E-03	2.3361E-03	1.3000E-03
67	976.20	1.4608E-08	8.2080E-01	3.0261E-03	2.3361E-03	1.3000E-03
68	977.21	1.4608E-08	8.3128E-01	3.0838E-03	2.3361E-03	1.3000E-03
69	978.22	1.4608E-08	8.4171E-01	3.1422E-03	2.3361E-03	1.3000E-03
70	979.23	1.4608E-08	8.5209E-01	3.2013E-03	2.3361E-03	1.3000E-03
71	980.24	1.4608E-08	8.6242E-01	3.2611E-03	2.3361E-03	1.3000E-03
72	981.25	1.4608E-08	8.7270E-01	3.3216E-03	2.3361E-03	1.3000E-03
73	982.26	1.4608E-08	8.8293E-01	3.3828E-03	2.3361E-03	1.3000E-03
74	983.27	1.4608E-08	8.9311E-01	3.4447E-03	2.3361E-03	1.3000E-03
75	984.28	1.4608E-08	9.0324E-01	3.5073E-03	2.3361E-03	1.3000E-03
76	985.29	1.4608E-08	9.1332E-01	3.5706E-03	2.3361E-03	1.3000E-03
77	986.30	1.4608E-08	9.2335E-01	3.6346E-03	2.3361E-03	1.3000E-03
78	987.31	1.4608E-08	9.3333E-01	3.6993E-03	2.3361E-03	1.3000E-03
79	988.32	1.4608E-08	9.4326E-01	3.7647E-03	2.3361E-03	1.3000E-03
80	989.33	1.4608E-08	9.5314E-01	3.8308E-03	2.3361E-03	1.3000E-03
81	990.34	1.4608E-08	9.6297E-01	3.8976E-03	2.3361E-03	1.3000E-03
82	991.35	1.4608E-08	9.7275E-01	3.9651E-03	2.3361E-03	1.3000E-03
83	992.36	1.4608E-08	9.8248E-01	4.0333E-03	2.3361E-03	1.3000E-03
84	993.37	1.4608E-08	9.9216E-01	4.1022E-03	2.3361E-03	1.3000E-03
85	994.38	1.4608E-08	1.0019E-01	4.1718E-03	2.3361E-03	1.3000E-03
86	995.39	1.4608E-08	1.0117E-01	4.2421E-03	2.3361E-03	1.3000E-03
87	996.40	1.4608E-08	1.0210E-01	4.3131E-03	2.3361E-03	1.3000E-03
88	997.41	1.4608E-08	1.0308E-01	4.3848E-03	2.3361E-03	1.3000E-03
89	998.42	1.4608E-08	1.0401E-01	4.4572E-03	2.3361E-03	1.3000E-03
90	999.43	1.4608E-08	1.0489E-01	4.5303E-03	2.3361E-03	1.3000E-03
91	1000.44	1.4608E-08	1.0582E-01	4.6041E-03	2.3361E-03	1.3000E-03
92	1001.45	1.4608E-08	1.0670E-01	4.6786E-03	2.3361E-03	1.3000E-03
93	1002.46	1.4608E-08	1.0763E-01	4.7538E-03	2.3361E-03	1.3000E-03
94	1003.47	1.4608E-08	1.0851E-01	4.8297E-03	2.3361E-03	1.3000E-03
95	1004.48	1.4608E-08	1.0944E-01	4.9063E-03	2.3361E-03	1.3000E-03
96	1005.49	1.4608E-08	1.1032E-01	4.9836E-03	2.3361E-03	1.3000E-03
97	1006.50	1.4608E-08	1.1125E-01	5.0616E-03	2.3361E-03	1.3000E-03
98	1007.51	1.4608E-08	1.1213E-01	5.1403E-03	2.3361E-03	1.3000E-03
99	1008.52	1.4608E-08	1.1306E-01	5.2197E-03	2.3361E-03	1.3000E-03
100	1009.53	1.4608E-08	1.1394E-01	5.2998E-03	2.3361E-03	1.3000E-03

GROUP 02  
 SYM PHIL X1  
 HFM TRANSFER PHASE  
 TRIM TEST  
 SYM PHIL X1

GROUP	MACH	RE NO	CONFIG	ALPHA	TIME	ASYMMETRIC		TRIM		TEST
						SYN	PHI	PHI	X/I	
37	B	1.25	X 10x6	0	0	0	0	0	0	225
				45		0	45			270
				90		0	90			315
				135		0	135			-
				180		0	180			-

2

ASYMMETRIC TRIM TEST

PHI X/I

1 2 3 4 5 6 7 8 9 10 11 12 13 14 15 16 17 18 19 20 21 22 23 24 25 26 27 28 29 30 31 32 33 34 35 36 37 38 39 40 41 42 43 44 45 46 47 48 49 50 51 52 53 54 55 56 57 58 59 60 61 62 63 64 65 66 67 68 69 70 71 72 73 74 75 76 77 78 79 80 81 82 83 84 85 86 87 88 89 90 91 92 93 94 95 96 97 98 99 100

0.00 0.20 0.40 0.60 0.80 1.00

0.00

0.10

0.20

0.30

0.40

0.50

0.1 x 10<sup>-4</sup>

0.1 x 10<sup>-4</sup>

0.00 0.10 0.20 0.30 0.40 0.50 0.60 0.70 0.80 0.90 1.00

X/L

0.00

0.20

0.40

0.60

0.80

1.00

0.00

0.20

0.40

0.60

0.80

1.00

0.00

0.20

0.40

0.60

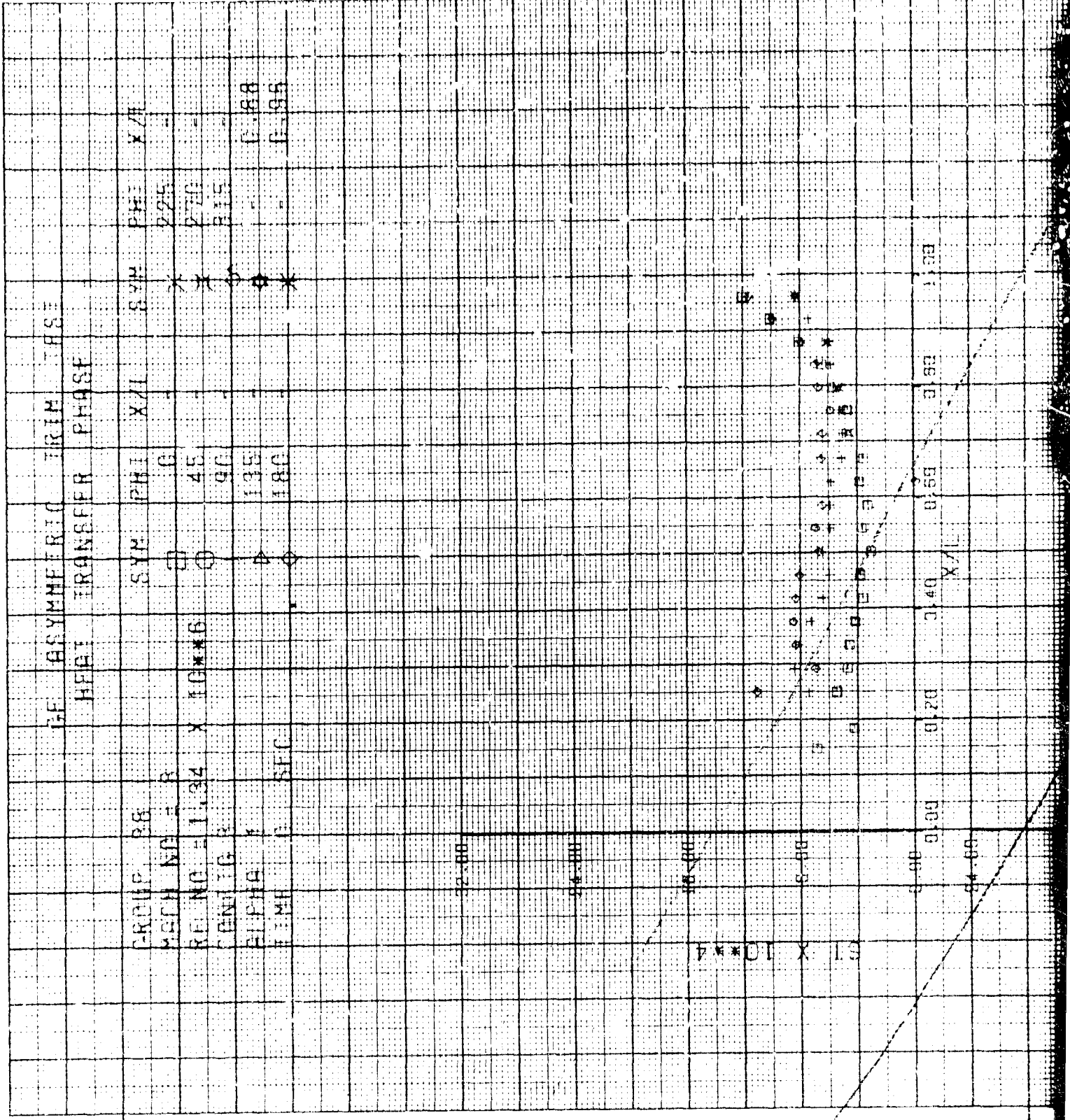
0.80

1.00

FIGURE 51

Figure 53

W



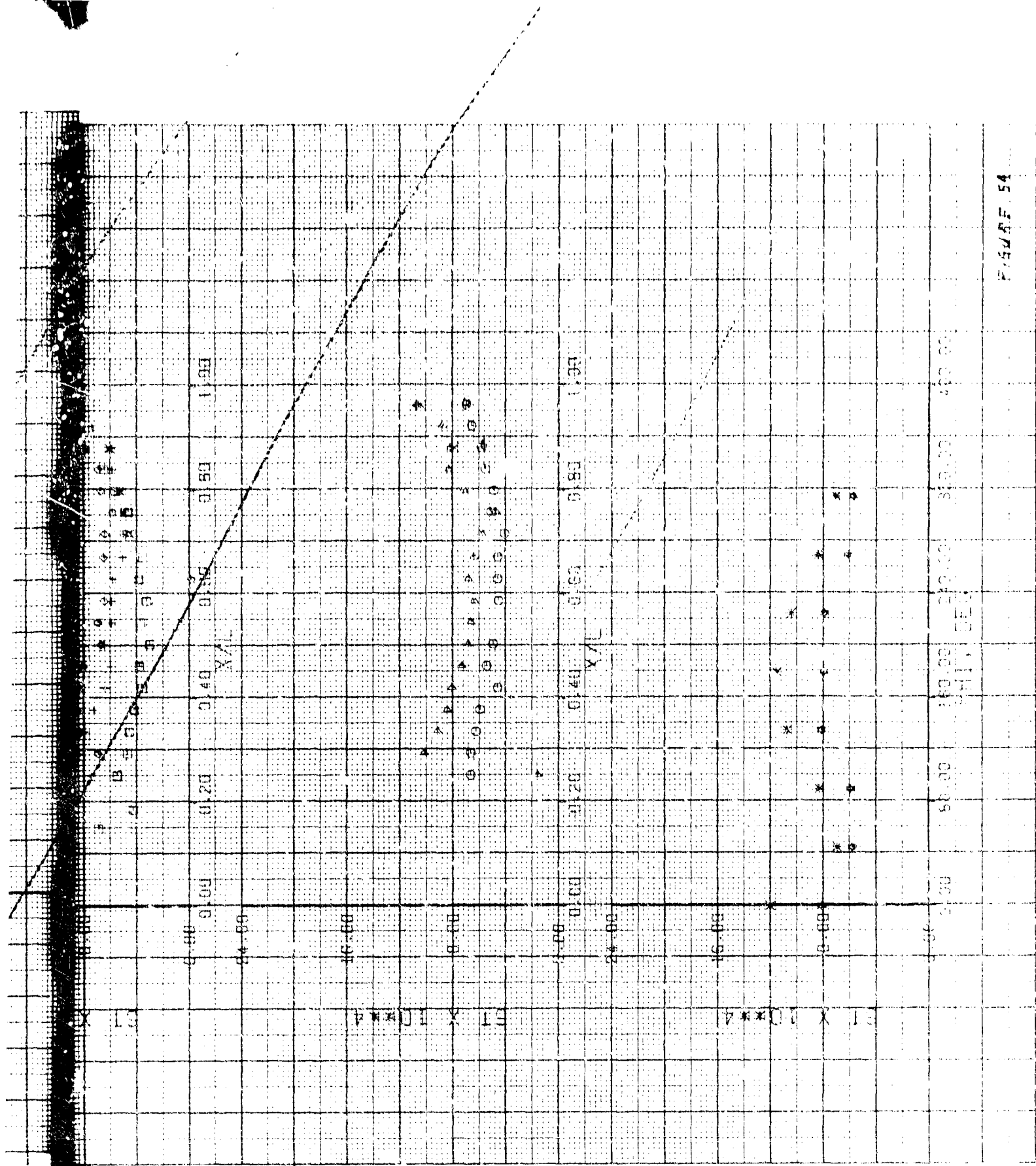


FIGURE 54

Figure 54

W

7/11/69

RESEARCH CENTER, AMMAN, JORDAN - JORDANIAN  
NORTH AMMAN GAS DYNAMICS FACILITY  
NO. 1000, HYPERSONIC TUNNEL

Table with columns: TEST NO., PRESSURE, MACH NO., REYNOLDS NO., ALPHA-ANGLE, HOLES, ALFA-ANGLE, DATE, MACH NO., PRESSURE, TO, TIME. Contains numerical data for various test runs.

GEOMETRIC RIN FIRST  
HEAT TRANSFER PHASE  
SYN. PHENYL SYM. PHT. XVI

GROUP	ASYNMETRIC	TRIP	TRIP	TRIP
NO	TRANSFER	PHASE	PHASE	PHASE
01				
02				
03				
04				
05				
06				
07				
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100				

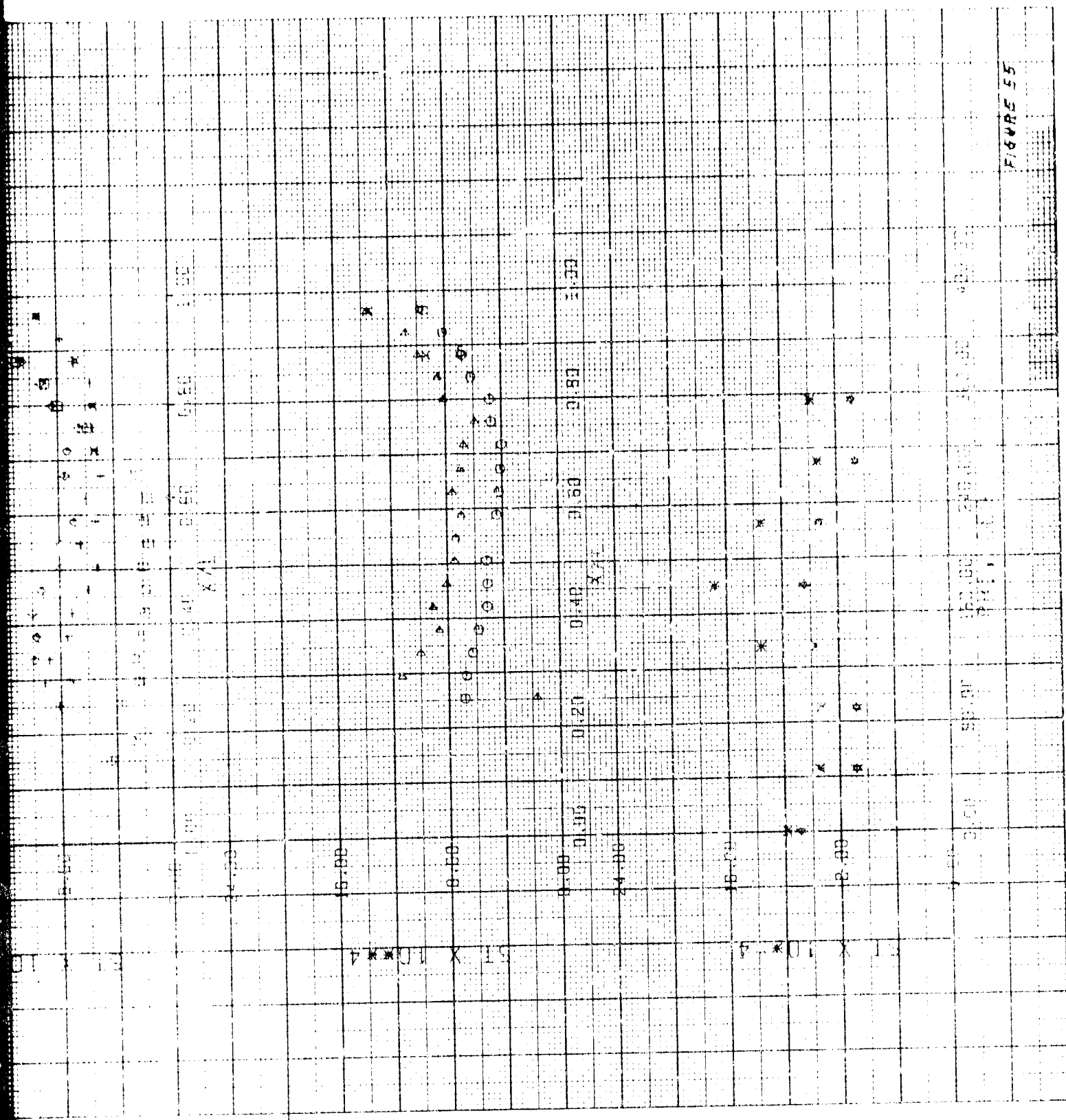


FIGURE 55

Figure 55

100

7/16/69

ARDEC (AND-1) - ANNEX D AF 51 TENNESSEE  
WALL HARMAN GAS DYNAMICS FACILITY  
50 INCH HYPERSONIC TUNNEL  
VI09AT-000

ALPHA-M BETA-M ALPHA-S MULT ALP-P DATE MACH NO PO TO TIME  
3 3.00 0 0 0 0 7.97 278.78 1257.05 1.00  
R-IMP 0.1000 0.1000 0.1000 0.1000 0.1000 0.1000 0.1000 0.1000 0.1000  
2.9200E-02 1.3010E-02 1.3010E-02 2.6410E-02 2.6410E-02 7.1300E-02 1.3500E-02 1.2024E-02 3.0701E-02

Table with columns: X, Y, PHI, TEMP, S(OPT), Q-SLOW, HT, ST. Rows contain numerical data for various parameters across different conditions.

THE SYMMETRIC KERN TEST  
HUE TRANSFER PHASE  
SYN PHIL XVI  
CIRCLE NO

GROUP NO	SYN	PHI	XI	SYM	PHI	XII
MACH NO 1-8		0			225	
REF NO 1-15 X 10005	□	45		X	270	
GENERIC 3	○	90		X	315	
RI-PHASE 4	△	135		○		0.88
TIME 0 SEC	◇	180		X		0.88

20

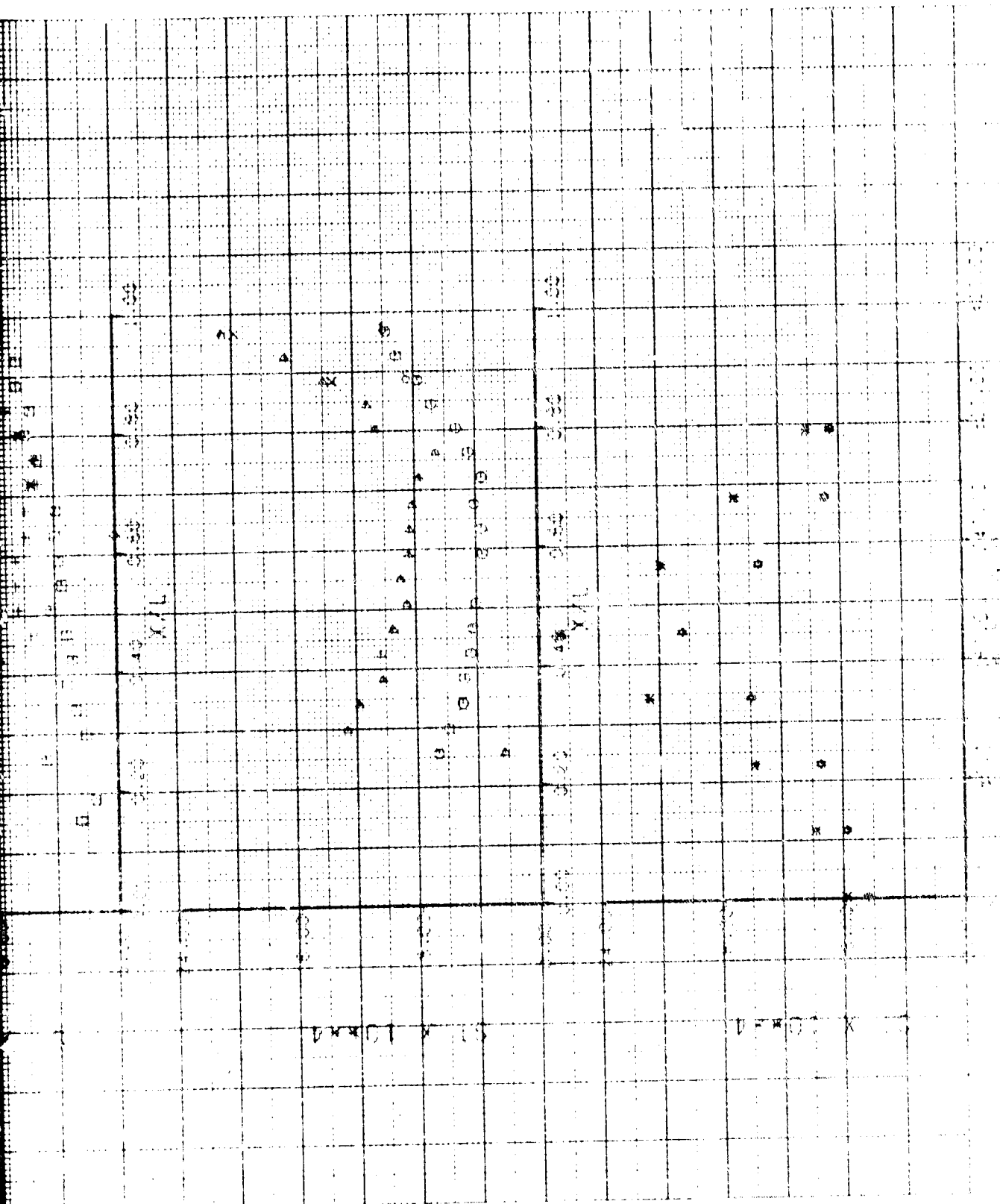


Figure 56

3





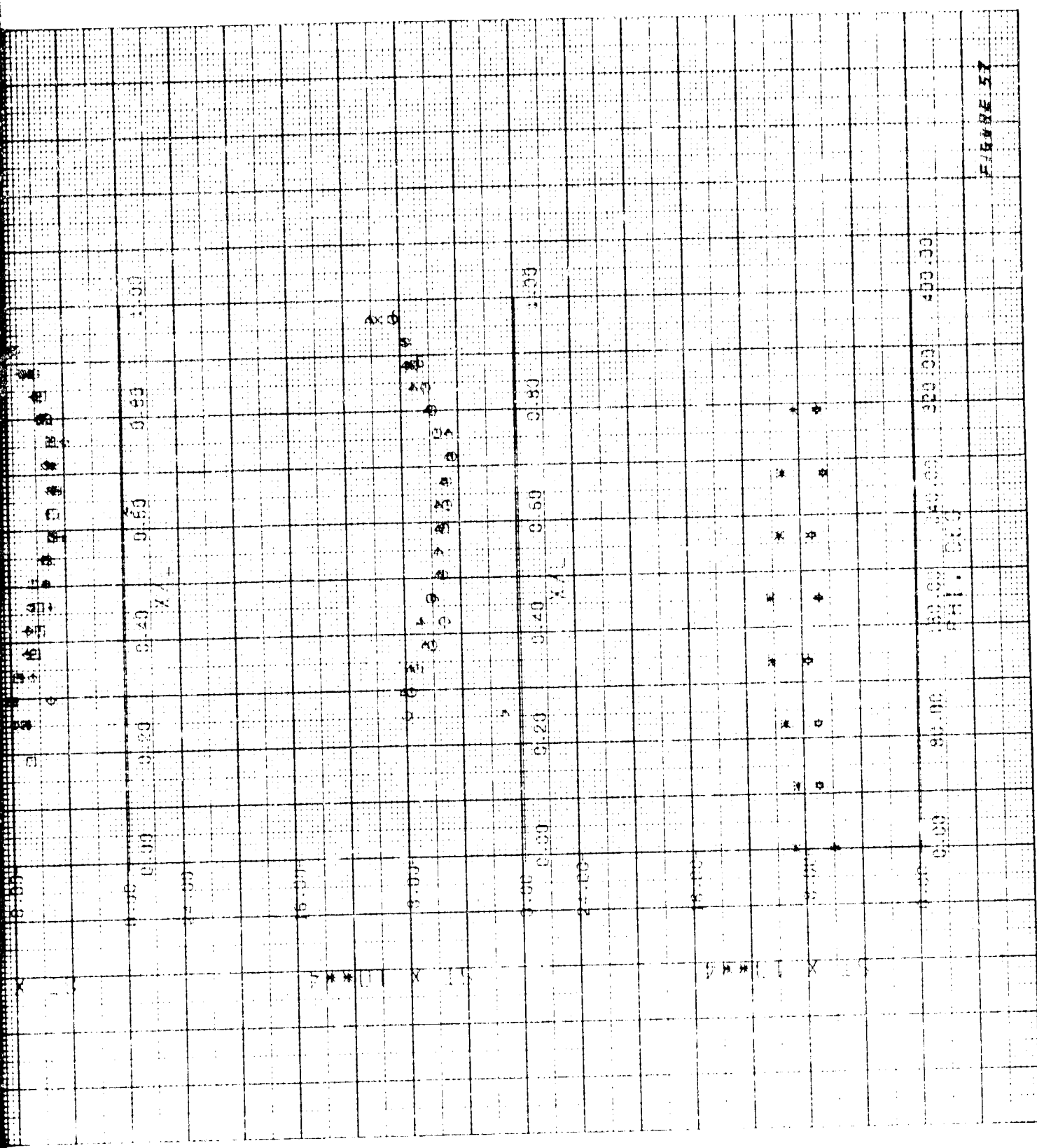


FIGURE 57

1.0

0.0

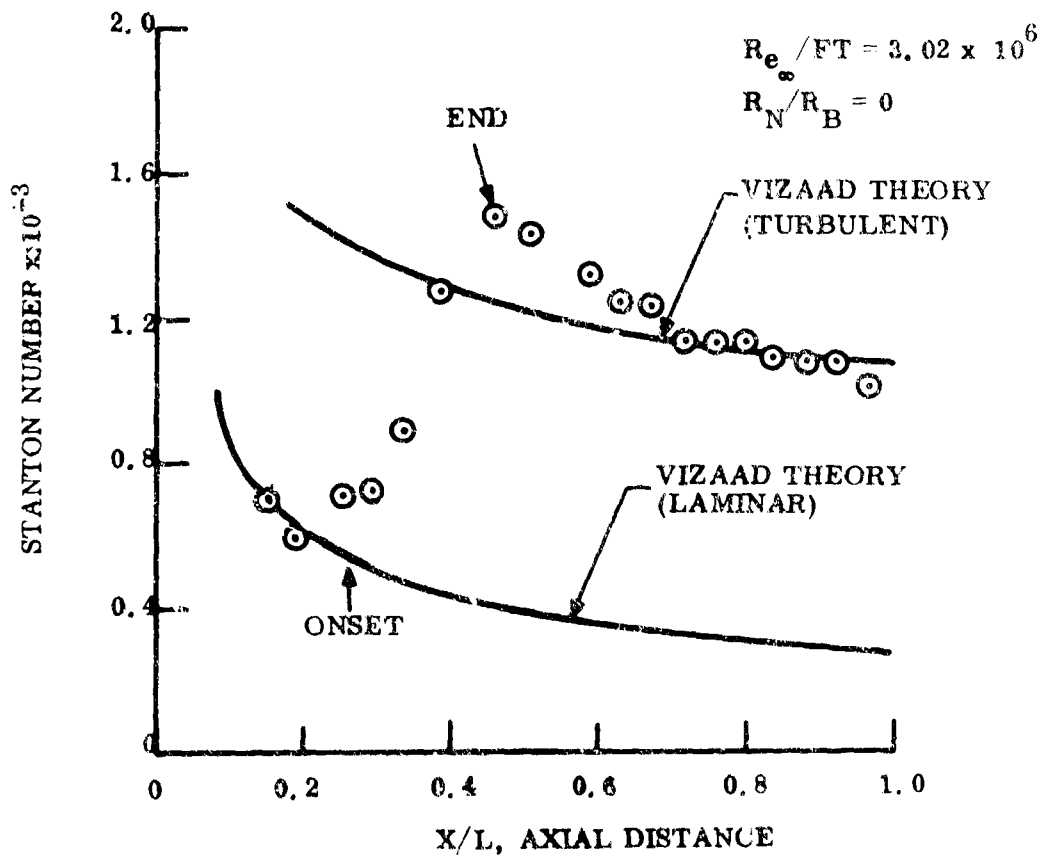
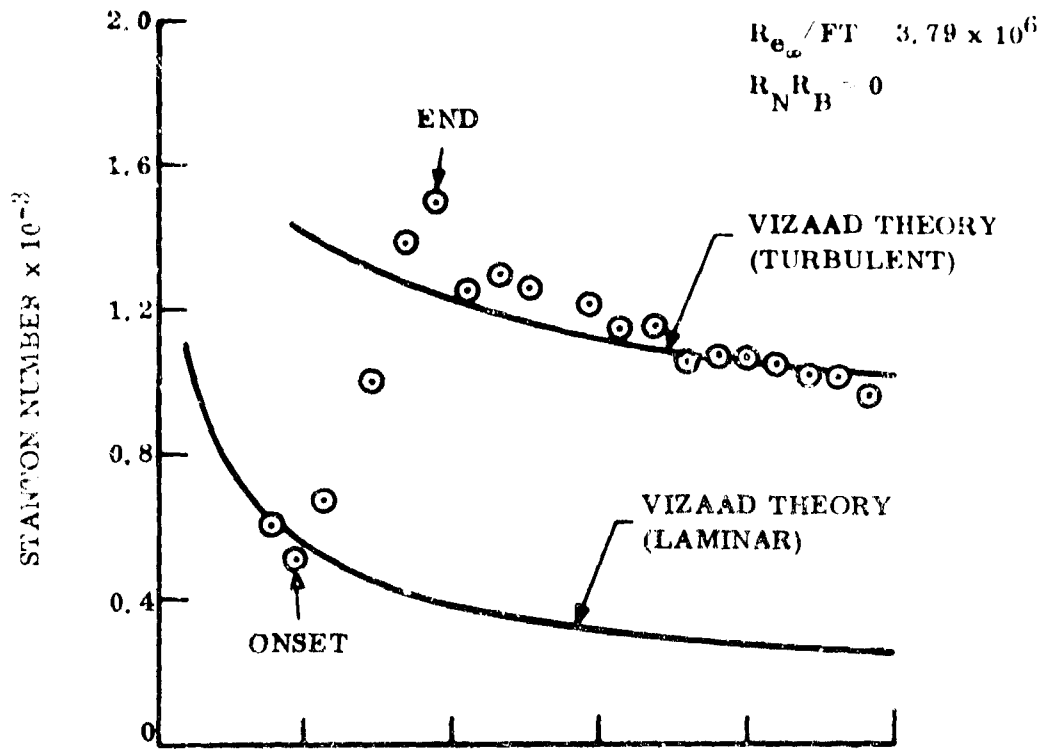


Figure 58. Stanton Number Distribution for a  $7.2^\circ$  Cone at Mach 8.  
 $Re_{\infty} = 3.79 \times 10^6$  per foot,  $R_N/R_B = 0$

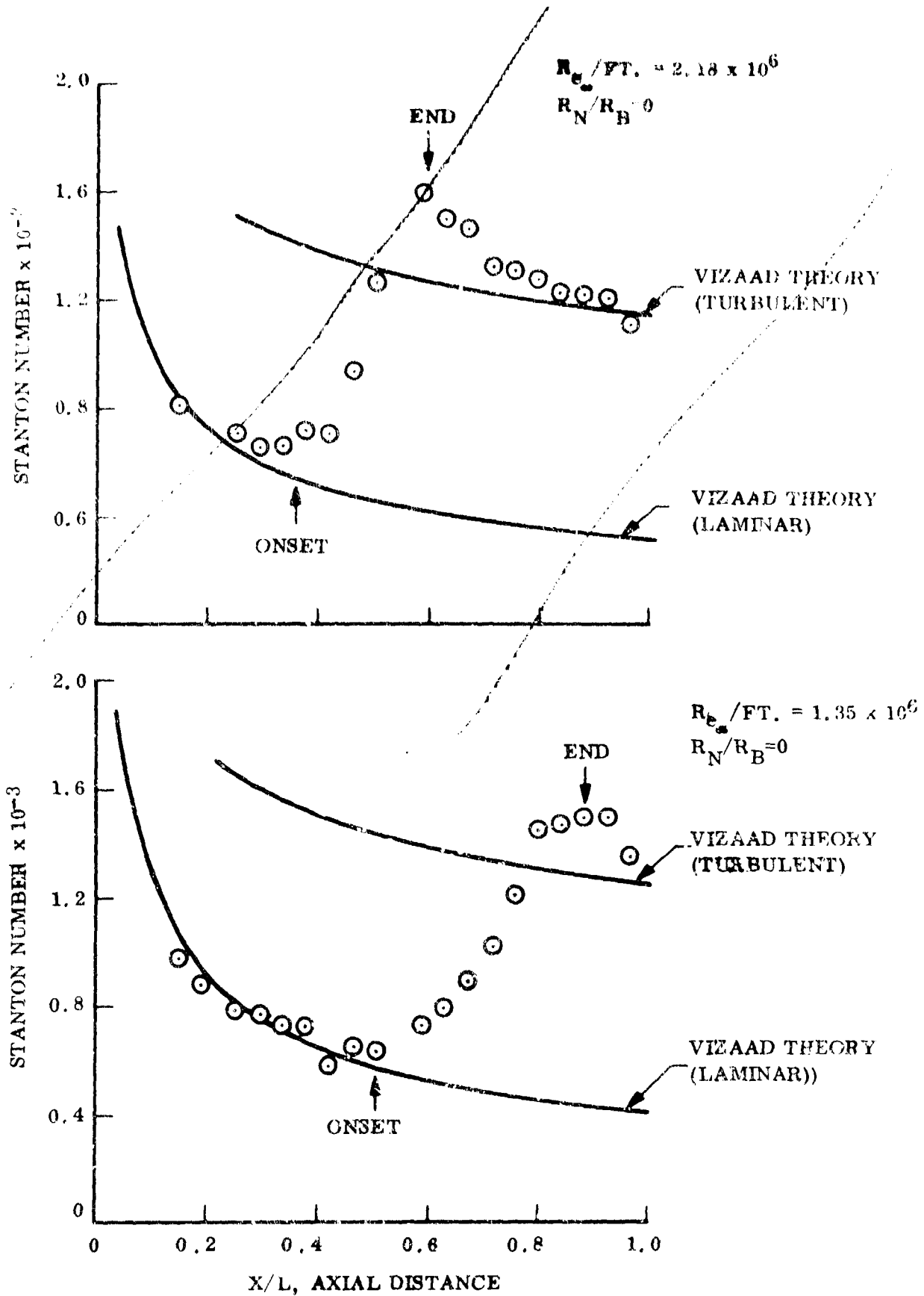


Figure 59. Stanton Number Distribution for a  $7.2^\circ$  Cone at Mach 8.  
 $Re_{\infty} 2.18 \times 10^6$  per foot,  $R_N/R_B = 0$

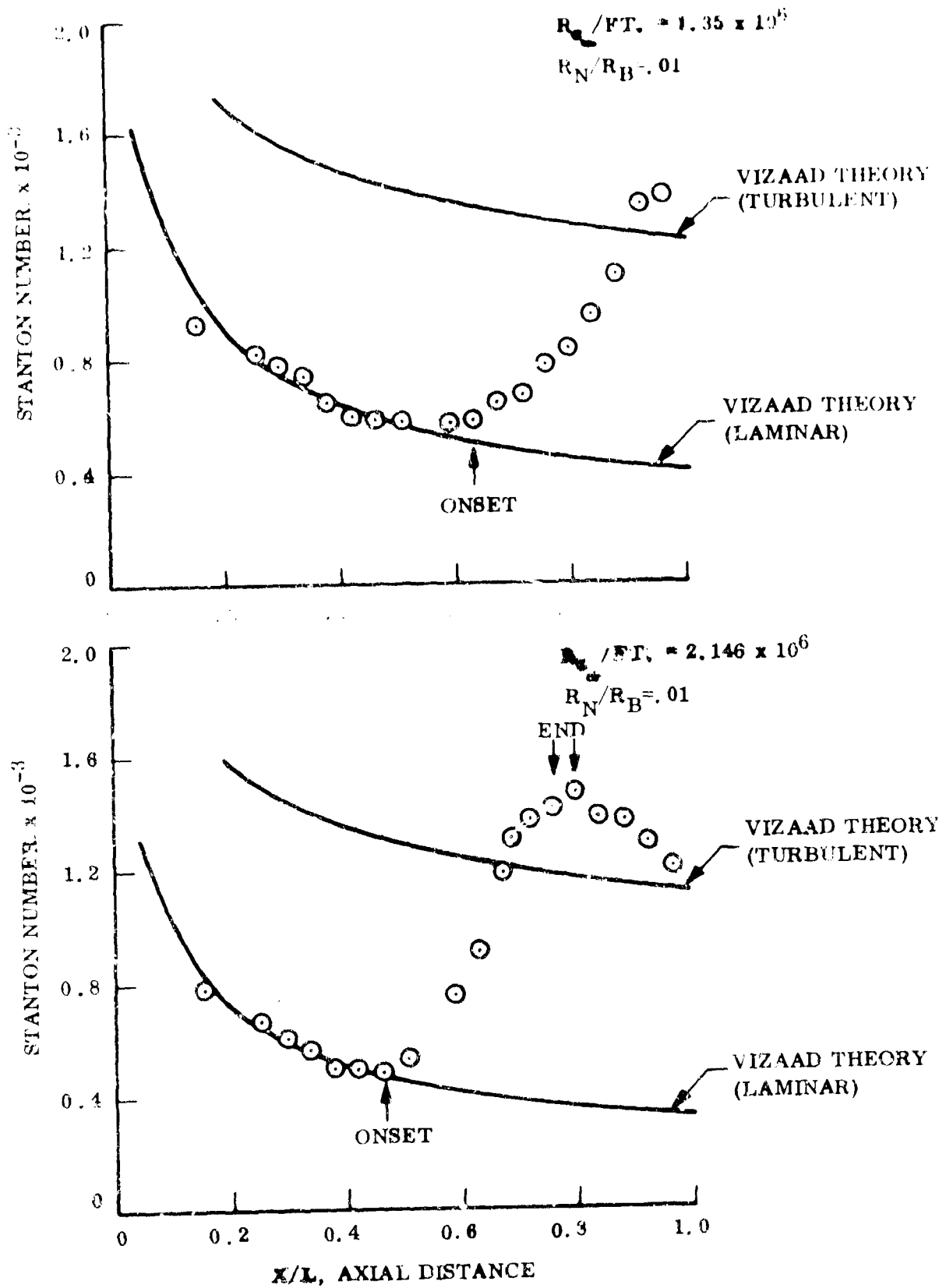


Figure 60. Stanton Number Distribution for a  $7.2^\circ$  Cone at Mach 8,  
 $Re_{\infty} 1.35 \times 10^6$  per foot,  $R_N/R_B = 0.01$

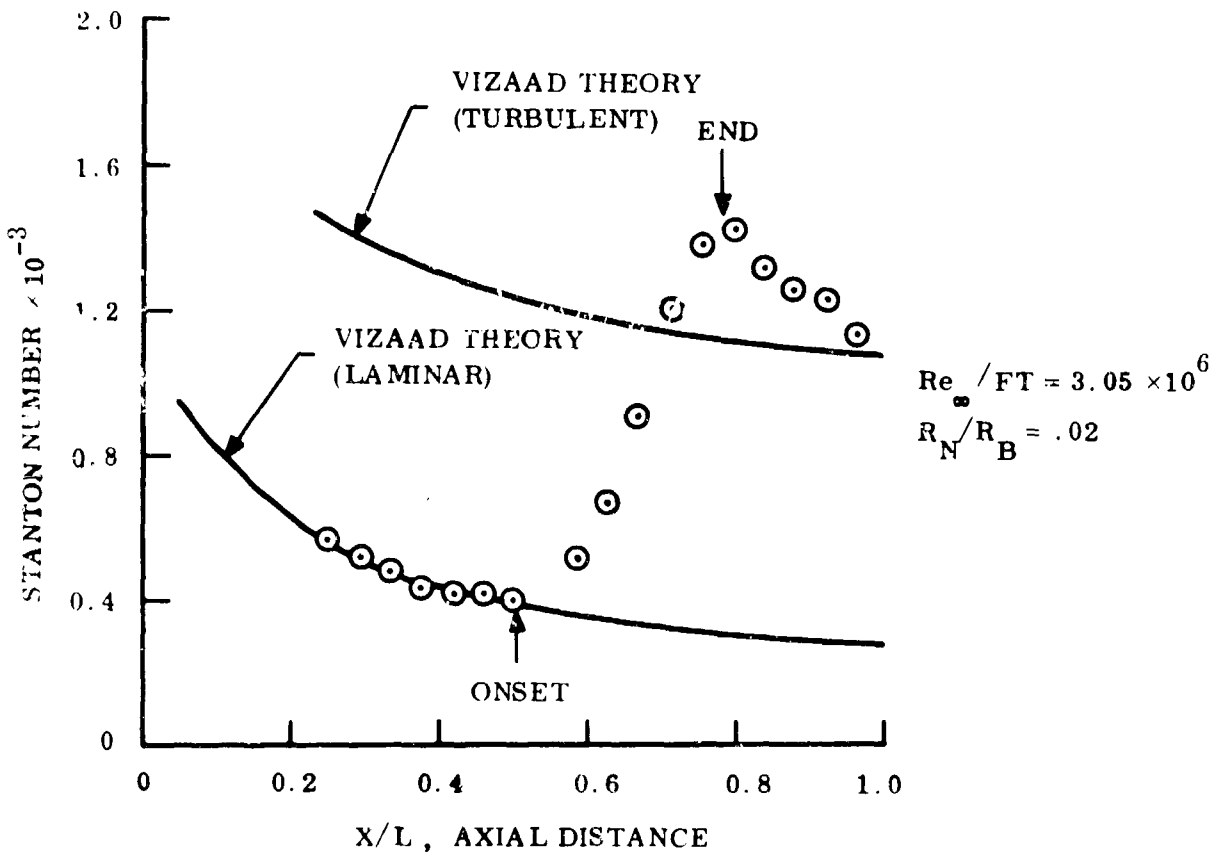
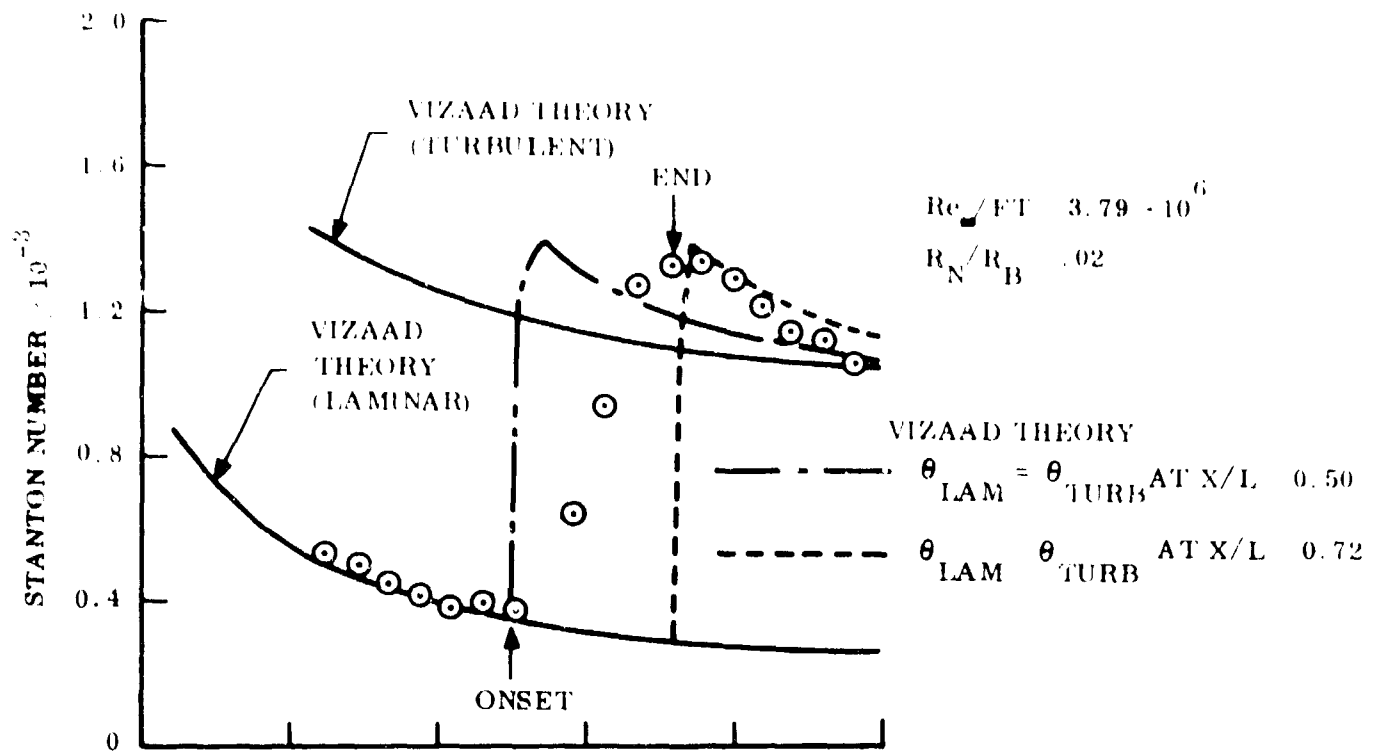


Figure 61. Stanton Number Distribution for a  $7.2^\circ$  Cone at Mach 8.  
 $Re_{\infty} = 3.79 \times 10^6$  per foot,  $R_N/R_B = 0.02$

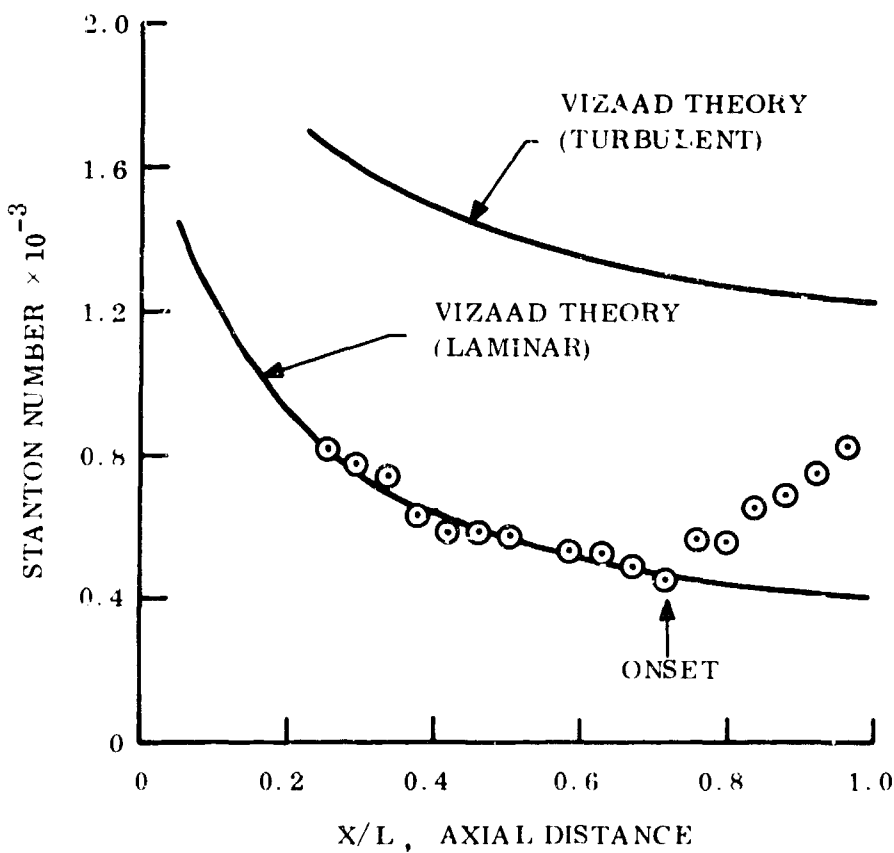
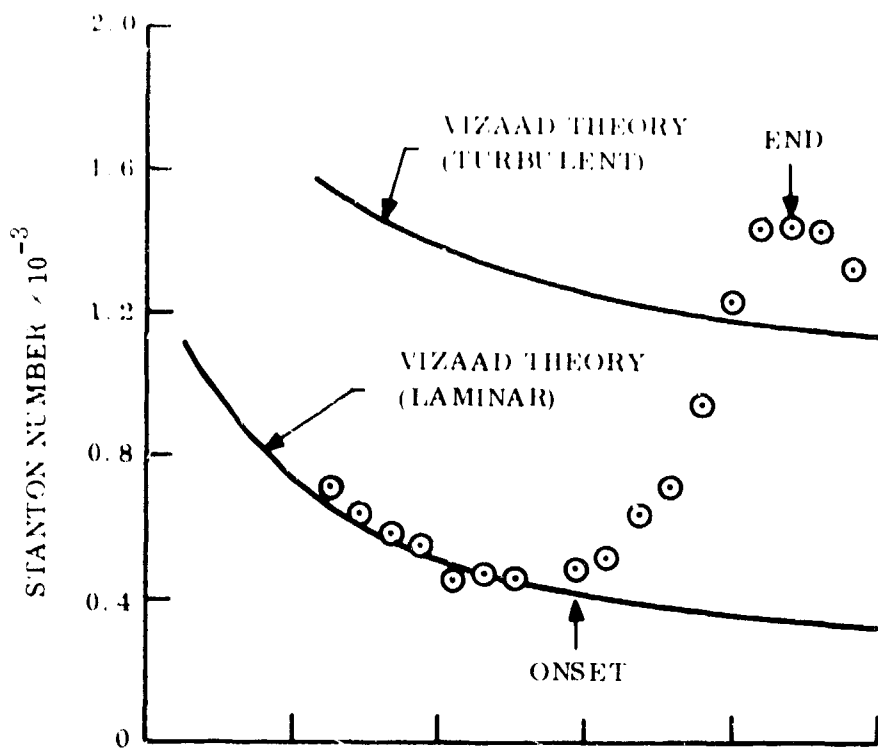


Figure 62. Stanton Number Distribution for a  $7.2^\circ$  Cone at Mach 8,  
 $Re_{\infty} = 2.20 \times 10^6$  per foot,  $R_N/R_B = 0.02$

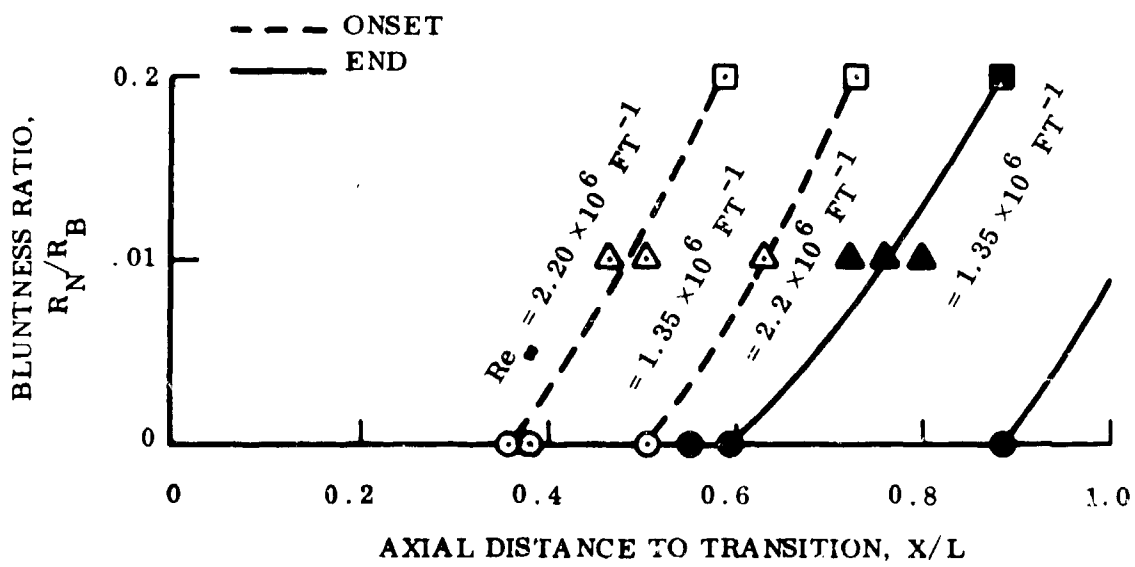
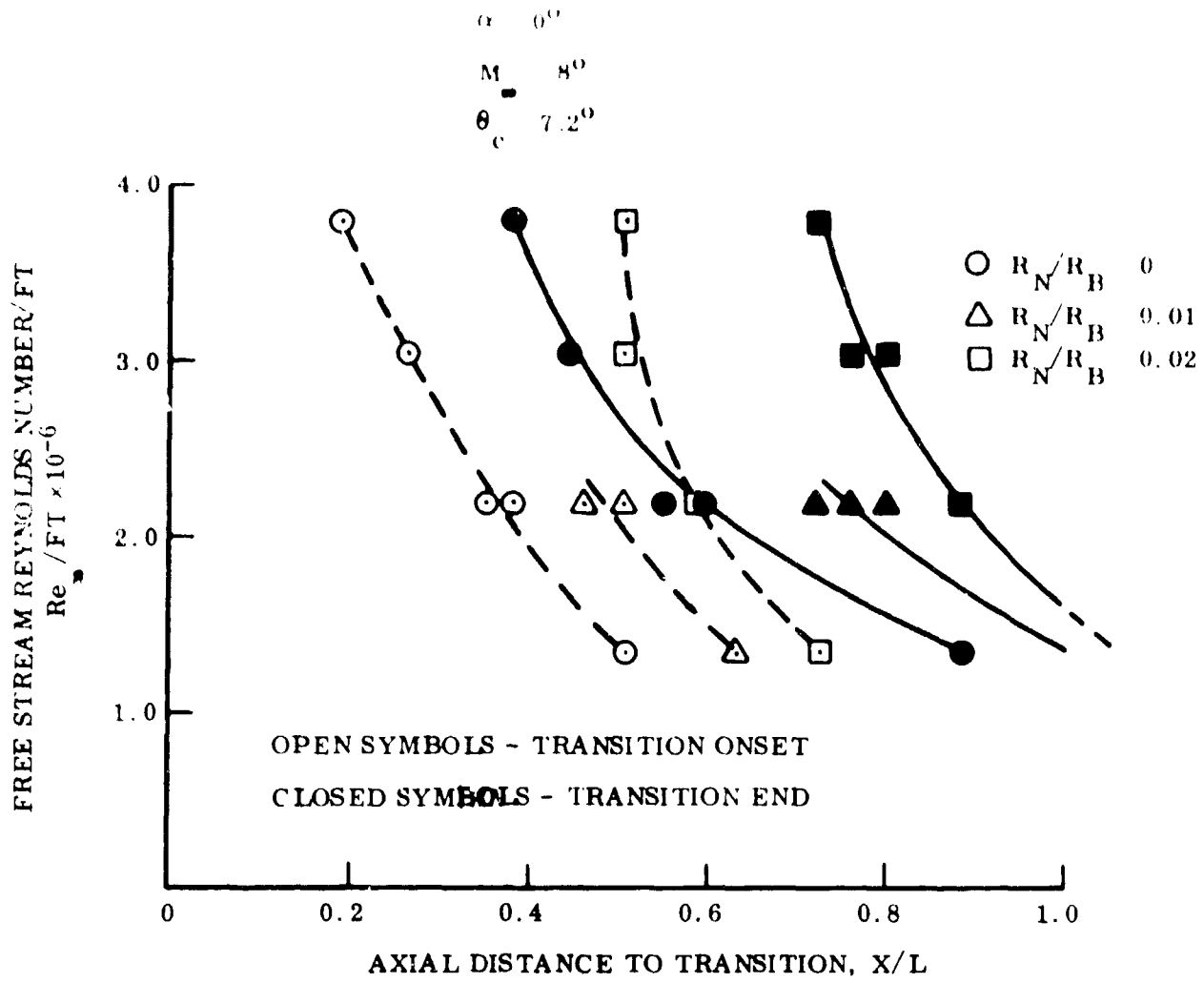


Figure 63. Effect of Reynolds Number and Bluntness on the Transition Region

$M_\infty = 8^\circ$   
 $\theta_c = 7.2^\circ$   
 $R_N/R_B = 0^\circ$

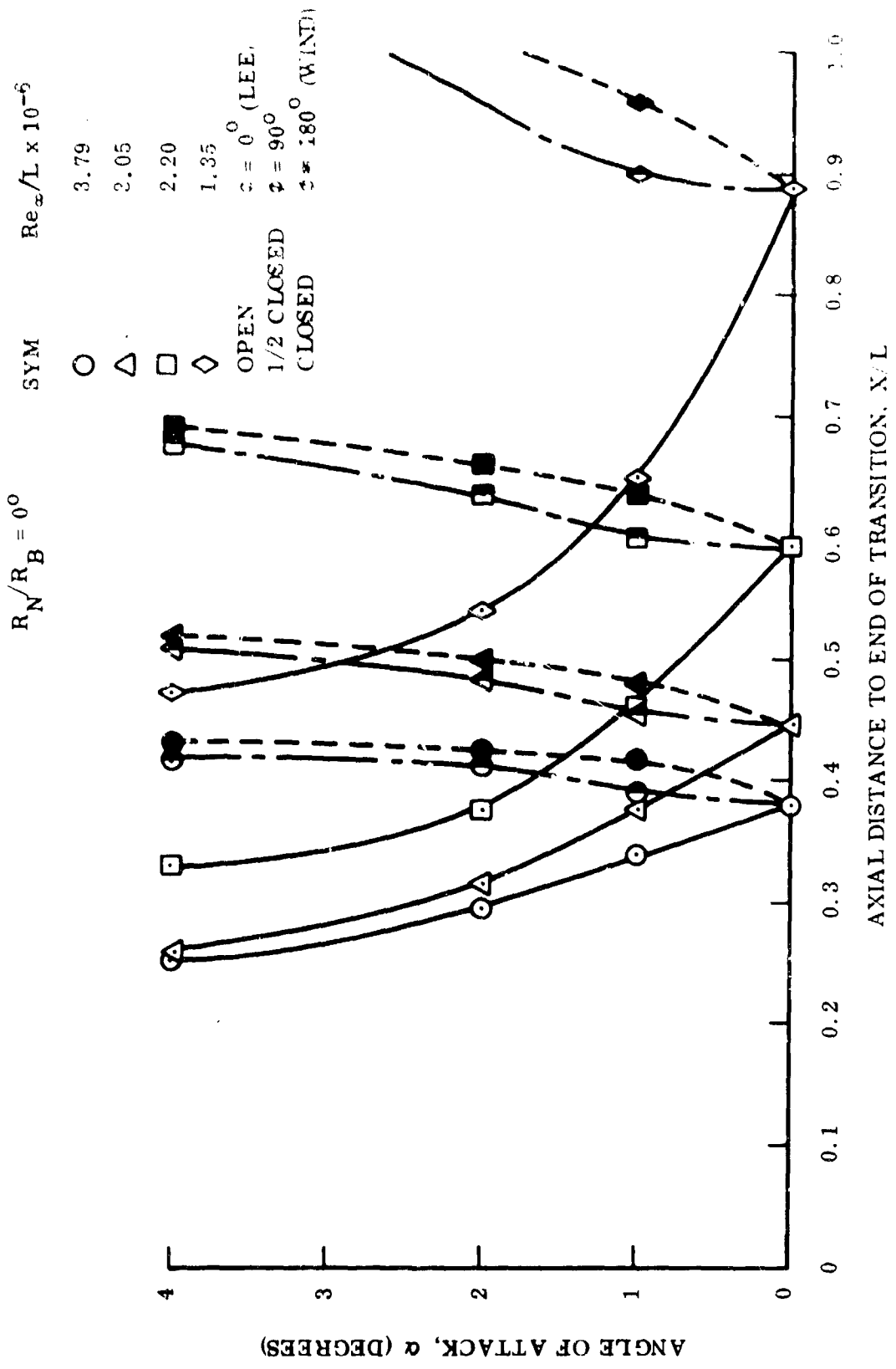


Figure 64. End of Transition Location on a Cone at Angle of Attack:  $M_\infty = 8$ ,  $\theta_c = 7.2^\circ$ ,  $R_N/R_B = 0$

$M_\infty = 8$   
 $\theta_c = 7.2^\circ$   
 $R_N/R_B = 0.01$

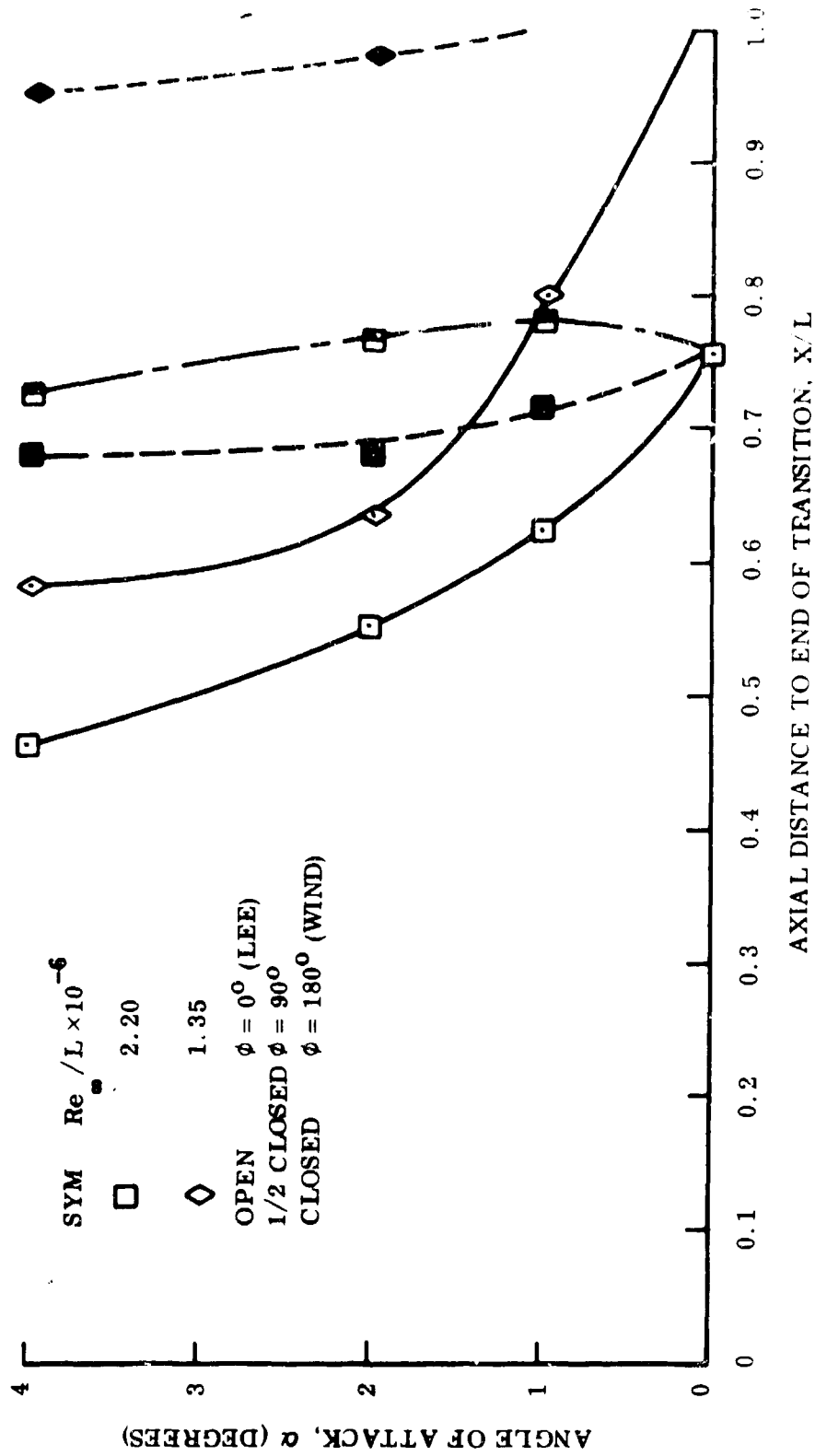


Figure 65. End of Transition Location on a Cone at Angle of Attack,  $M_\infty = 8$ ,  $\theta_c = 7.2^\circ$ ,  $R_N/R_B = 0.01$

$M_\infty = 8$   
 $\theta_c = 7.2^\circ$   
 $R_N/R_B = 0.02$

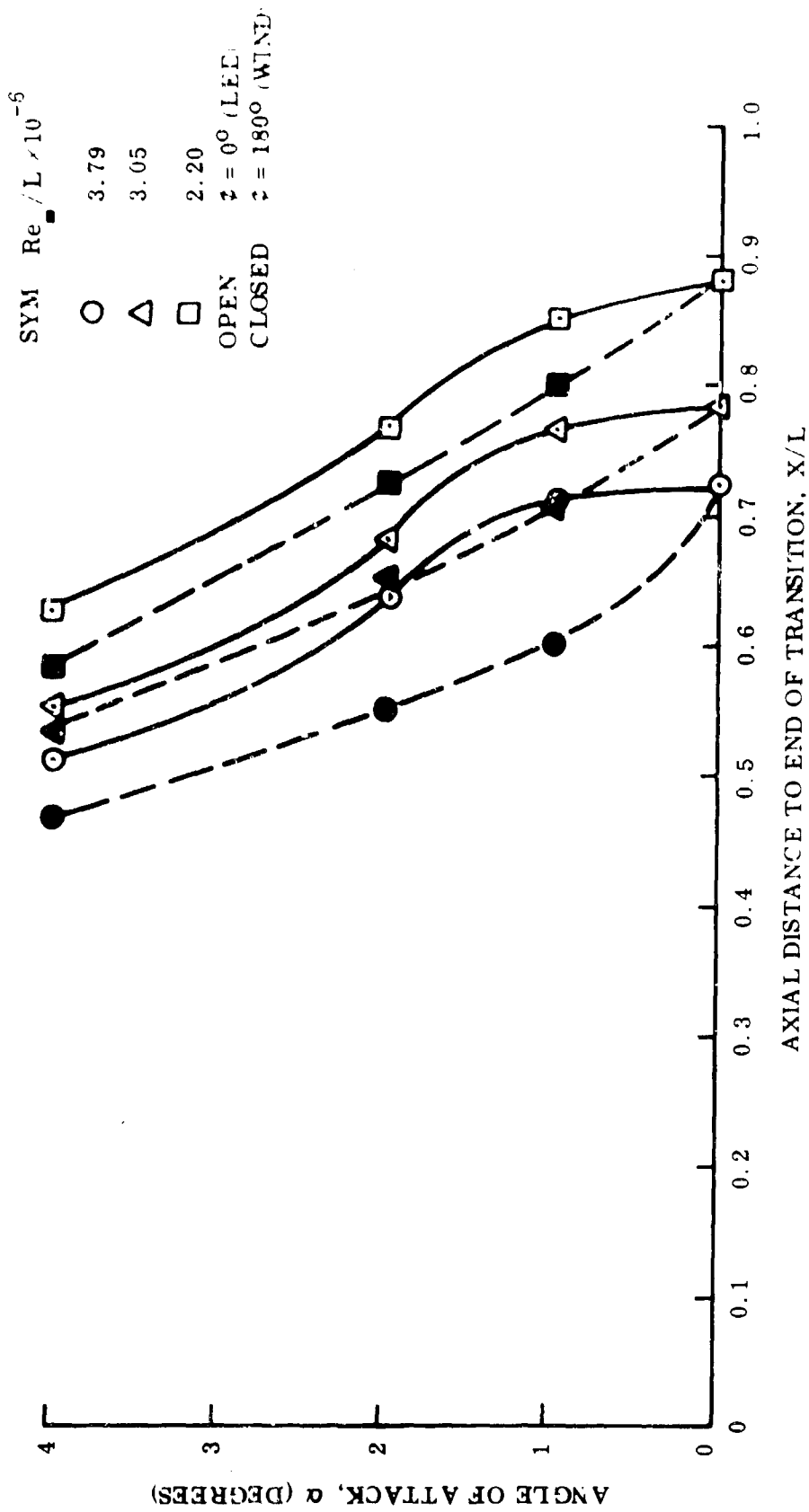


Figure 66. End of Transition Location on a Cone at Angle of Attack,  $M_\infty = 8$ ,  $\theta_c = 7.2^\circ$ ,  $R_N/R_B = 0.02$

$M_\infty = 8$   
 $R_N/R_B = 0$   
 $\theta_c = 7.2^\circ$

$X/L = 1.0$

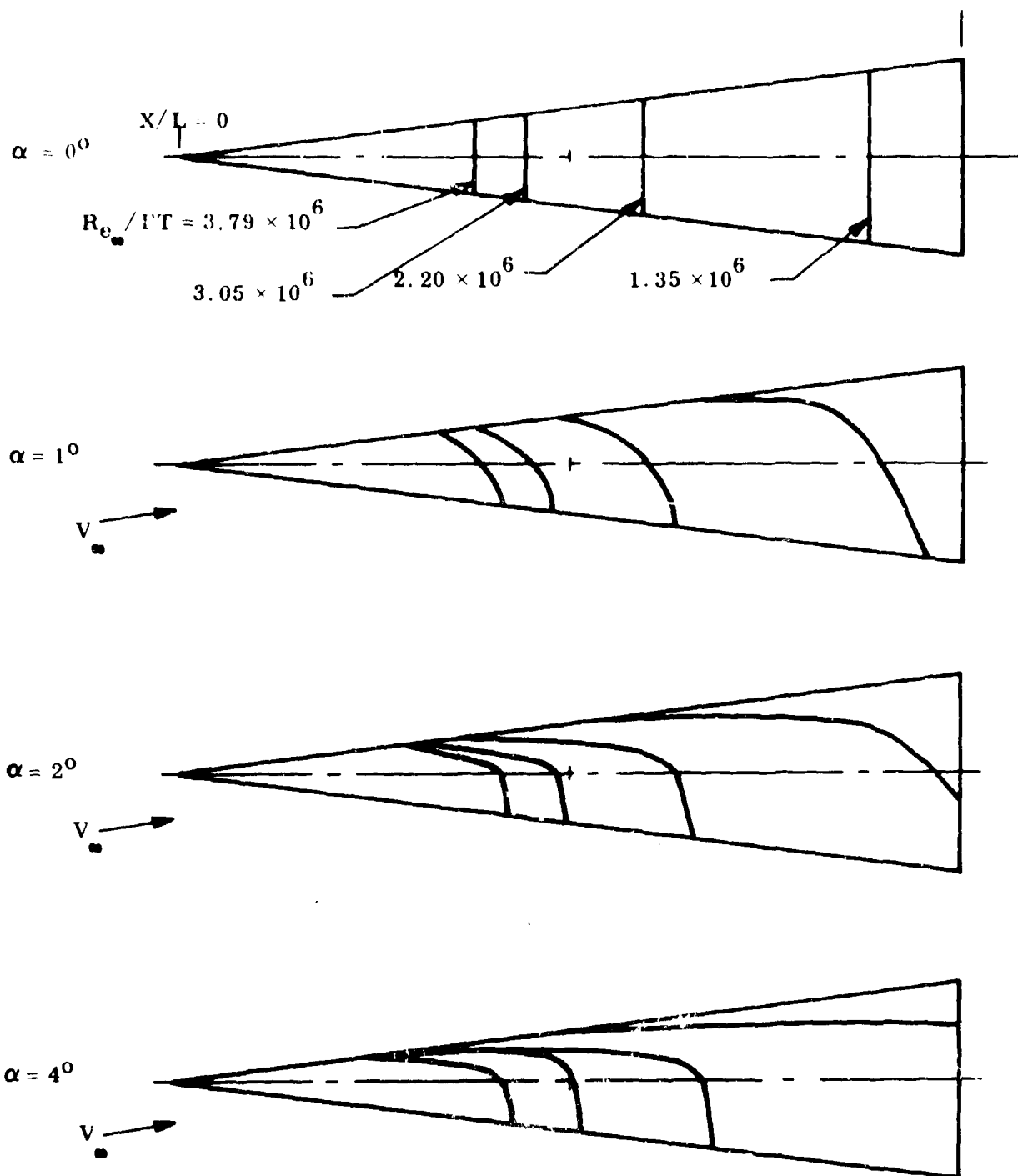


Figure 67. Spatial Distribution of End of Transition with Reynolds Number and Angle of Attack,  $M_\infty = 8$ ,  $R_N/R_B = 0$ ,  $\theta_c = 7.2^\circ$

$M_\infty = 8$   
 $R_N/R_B = 0.01$   
 $\theta_c = 7.2^\circ$

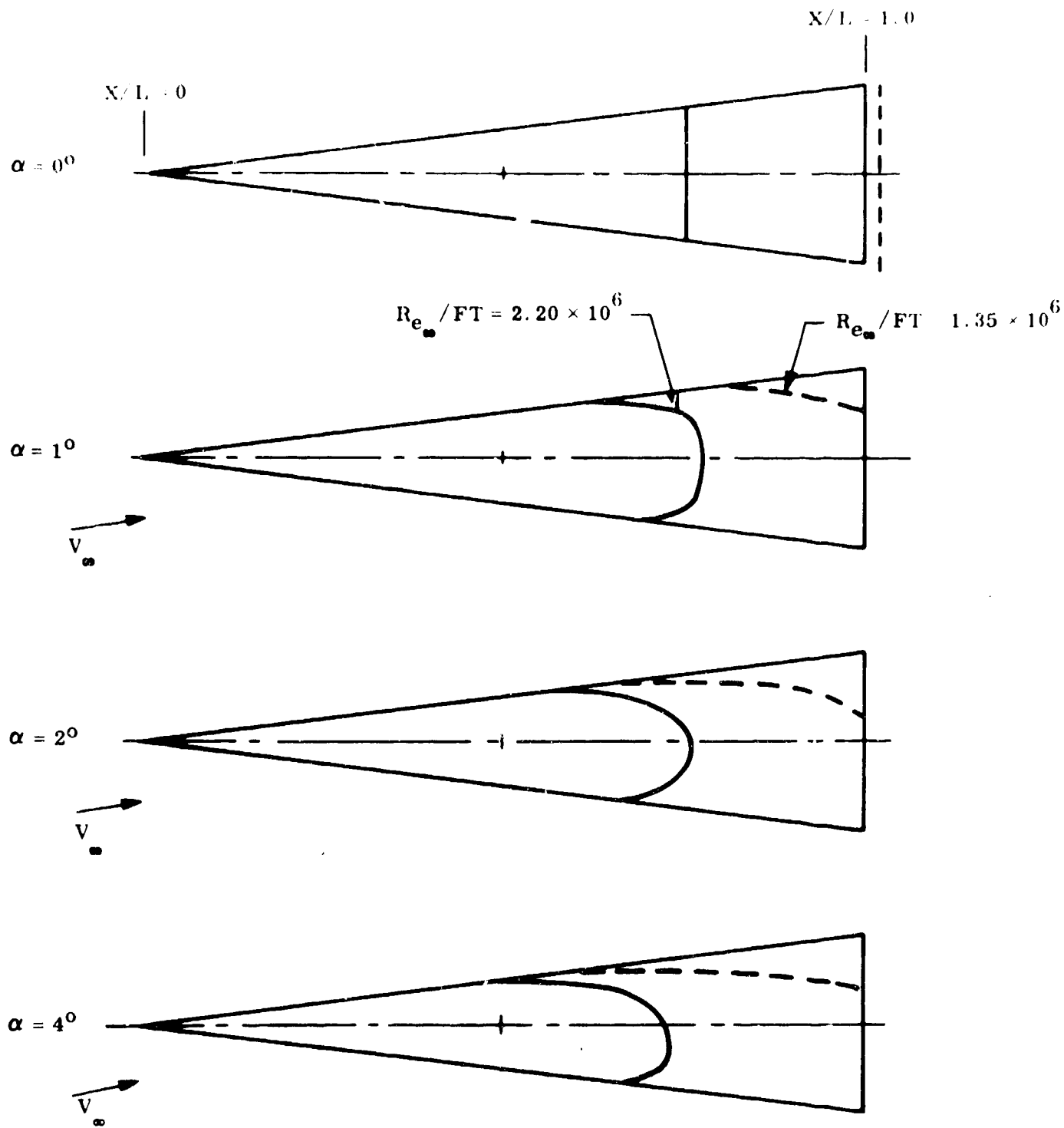


Figure 68. Spatial Distribution of End of Transition with Reynolds Number and Angle of Attack.  $M_\infty = 8$ ,  $R_N/R_B = 0.01$ ,  $\theta_c = 7.2^\circ$

$$M_\infty = 8$$

$$R_N/R_B = 0.02$$

$$\theta_c = 7.2^\circ$$

X/L = 1.0

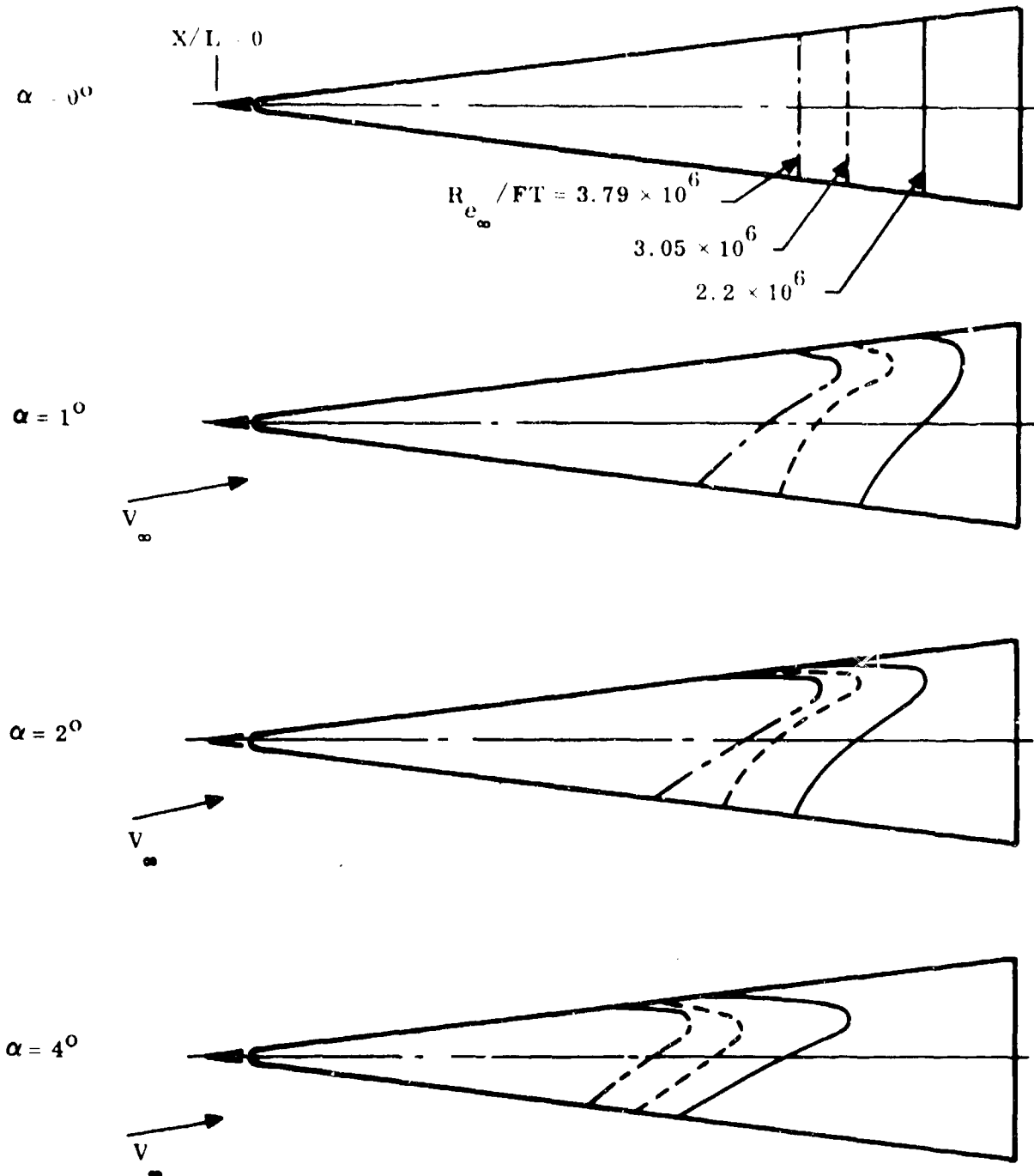
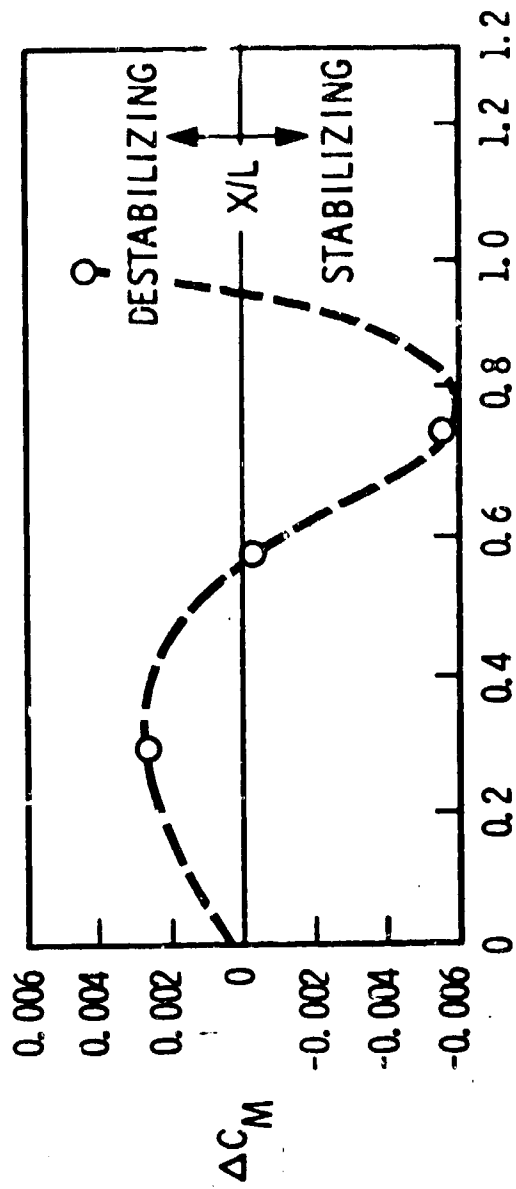


Figure 69. Spatial Distribution of End of Transition with Reynolds Number and Angle of attack,  $M_\infty = 8$ ,  $R_N/R_B = 0.02$ ,  $\theta_c = 7.2^\circ$



$M_{\omega} = 8.0$   
 $\bar{m} = 0.025$   
 $\alpha = 1^{\circ}$

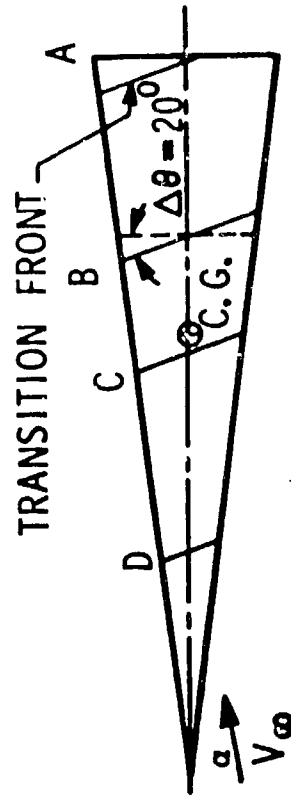


Figure 70. Effect of Asymmetric Transition on Pitching Moment Coefficient

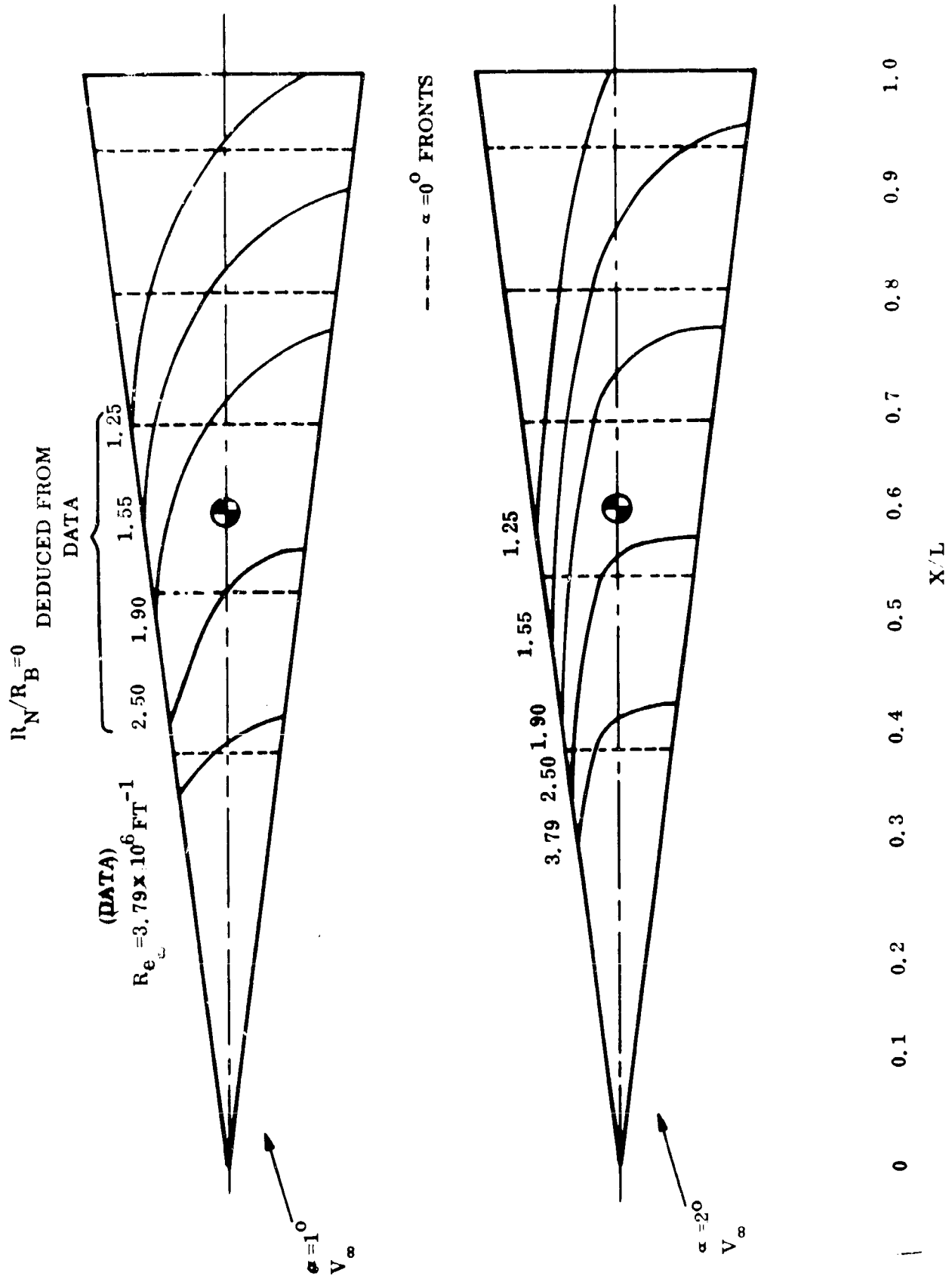


Figure 71. Proposed End of Transition Front Locations for Force Tests.  $R_N/R_B = 0$

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4. DESCRIPTIVE NOTES (Type of report and inclusive dates) Technical - second half of CY 1969		
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14 KEY WORDS	LINK A		LINK B		LINK C	
	ROLE	WT	ROLE	WT	ROLE	WT
Hypersonic Flow						
Boundary Layer Transition						
Heat Transfer						
Angle of Attack						
Bluntness						
Reynolds Number						

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