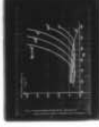
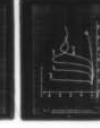
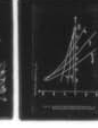
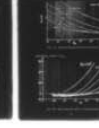
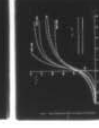
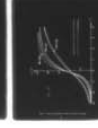
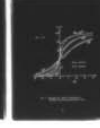
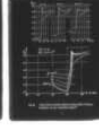


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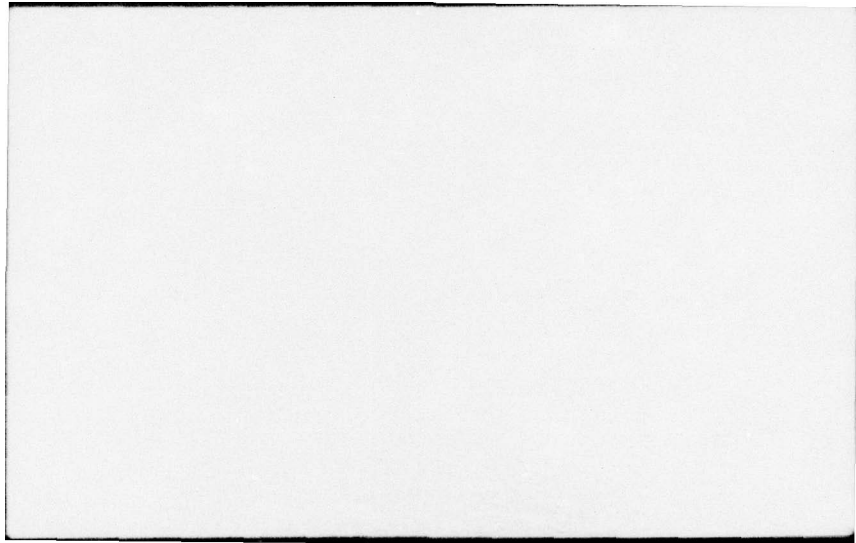
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SHOCK WAVE PENETRATION AND LATERAL PRESSURE
GRADIENT EFFECTS ON TRANSONIC NORMAL
SHOCK-TURBULENT BOUNDARY LAYER
INTERACTIONS†

G. R. INGER*

December 1976

† Based on research partially supported by the U.S. Office of Naval Research under Contract N00014-75-C-0456 and by the Alexander Von Humboldt Foundation.

* Professor of Aerospace and Ocean Engineering. Von Humboldt Visiting Senior Research Awardee, DFVLR-AVA, Göttingen, West Germany (1976-1977)

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NOMENCLATURE

| | |
|---------------|--|
| C_f | local skin friction |
| h_i | P_i'/P_{0i} ($i = 1, 2, 3$) |
| H_i | Fourier transform of h_i |
| k | wave number in Fourier Transform |
| δ | sublayer thickness (see \rightarrow Eq. (6)) |
| p | static pressure |
| Q | unit function defined in Eq. (4) |
| Re | Reynolds number based on L |
| u, v | x- and y- velocity components |
| x, y | coordinates parallel and perpendicular to freestream |
| β | $(M^2 - 1)^{1/2}$ or $(1 - M^2)^{1/2}$ |
| γ | ratio of specific heats |
| δ_{BL} | boundary layer thickness |
| ΔP | basic static pressure jump across shock |
| η | displaced interface location |
| μ | viscosity coefficient |

- π Fourier Transform of pressure disturbance function
- ρ density
- ω exponent for power law viscosity law ($\mu \sim T^\omega$)

Superscripts

- ()' disturbance quantity

Subscripts

- e freestream properties upstream of shock at boundary layer edge
- o denotes undisturbed (not stagnation) flow property
- L length to undisturbed shock location
- w property at wall value
- i $i = 1, 2, 3$, denotes region of variable
- LS based on Lighthill sublayer

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ABSTRACT

The titled affects, which are usually neglected in contemporary theories of viscous-inviscid interaction, are here examined in detail for the unseparated transonic turbulent case. They are found to be significant in many respects. In particular, our results suggest that neglect of shock penetration becomes very inaccurate in the supersonic separated flow regime, leading to substantial underestimates of the local interact strength and its overall stream-wise extent.

SHOCK WAVE PENETRATION AND LATERAL PRESSURE
GRADIENT EFFECTS ON TRANSONIC NORMAL SHOCK-
TURBULENT BOUNDARY LAYER INTERACTIONS⁺

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INTRODUCTION

Basic studies of shock wave-turbulent boundary layer interaction find significant practical application in various aerodynamic and propulsion system flow problems. In existing theories the impinging shock is imposed as an outer disturbance boundary condition at the boundary layer edge but the local details of its subsequent penetration into the layer neglected in obtaining the interactive flow field solution. Moreover, the attendant simplification of neglecting the lateral interaction-pressure gradient across the boundary layer is often made. These approximations are usually justified a priori on the grounds that the resulting errors in the important large scale features of the flow are insignificant for engineering purposes. Indeed, up to moderate supersonic speeds this has been found true in laminar flows because of their well-spread out response to even weak shocks and the small attendant lateral pressure variations across the boundary layer. However, for turbulent flows at finite Reynolds numbers of practical interest ($10^6 \leq Re_L \leq 10^8$) the interaction is much more violent and short range¹ ($\pm 2 - 4 \delta$) and

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Based on research partially supported by the U. S. Office of Naval Research under contract N 00014-75-C-0456 and by the Alexander Von Humboldt Foundation. Helpful discussions with Drs. Mike Werle and A. Kluwick are acknowledged. Appreciation is also extended to Mr. S. Zee for his assistance with the calculations.

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hence the shock penetration and lateral pressure gradient effects may become important. Werle and Bertke² have recently given strong evidence that this is the case: detailed comparisons show that their interacting supersonic boundary layer model solutions (which have otherwise given consistently good results in laminar flows) severely misrepresent both experimental data and exact Navier-Stokes solutions³ for a $M_\infty = 3$ separating $Re_L = 10^7$ turbulent flow regardless of the eddy viscosity model employed. They concluded that the failure of the model is due to its lack of account for the shock penetration into the boundary layer; moreover, neglected lateral pressure gradient terms may also play a significant role in such cases.

The present paper deals with these aspects of the interaction problem for the case of transonic normal shocks interacting with non-separating turbulent boundary layers. Although acknowledged to be in a lower speed range without separation the results nevertheless shed valuable light on the nature and parametric dependence of the shock penetration and $\partial p/\partial y$ effects which proves useful in other situations. The approach is based on an extension of a recently-developed approximate theory⁴ of this transonic interaction problem.

THEORETICAL MODEL

The flow is taken to consist of a known unseparated turbulent boundary layer profile $M_0(y)$ subjected to small transonic disturbances due to an impinging weak normal shock ($M_1 < 1.3$). Our original theoretical model of this interaction was an approximate simplification of the disturbance flow structure emerging from an asymptotic analysis of the compressible Navier-Stokes equations at high Reynolds numbers, giving a linearized boundary value problem surrounding the nonlinear shock discontinuity and underlaid by a thin Lighthill viscous disturbance sublayer (Fig. 1). Provided the incident shock is weak enough to avoid a separated lambda-shock interaction pattern⁵ while M_1 is not so close to unity that the viscous structure within the incident shock must be considered, this flow model represents all the essential

global features of the mixed transonic character of the non-separating normal shock-boundary layer interaction problem⁶ including lateral pressure gradient effects. Moreover, the linearized theory involved is amenable to analytical treatment by obtaining solutions in each of the three regions shown in Fig. 1 and matching them along the boundary layer edge interface by continuity of pressure and streamline slope. The details, carried out by operational methods, are given elsewhere⁴. The results include interaction pressure and streamline deflection along the boundary layer edge and both the wall pressure and shear distributions over a wide range of M_1 and Re_L values for an arbitrary incoming turbulent boundary layer profile, whether adiabatic or not⁷.

As originally formulated, the aforementioned theory not only assumes that the incident shock and its extension down to the sonic line are simple Rankine-Hugoniot discontinuities, but also neglects (following Lighthill's treatment⁸ of oblique shock interaction) the detailed shock penetration into the boundary layer by imposing the shock jump conditions only at the outer edge; since the correct shock pressure jump at the edge is accounted for while below the sonic level in the boundary layer no discontinuity can exist, the shock decay across the supersonic non-uniform flow region is in fact roughly simulated by this approximation. The ultimate result is a small wall pressure discontinuity under the shock (Fig. 2) whose effect can be easily eliminated by an equivalent continuous redistribution of the jump over the entire interaction⁹. Nevertheless, since it is of fundamental and practical interest to improve on this approximation, a modification of the theory to account for the shock penetration effect was sought.

An earlier preliminary study⁹ has shown that except very near the sonic level y_s , the shock in a turbulent boundary layer remains essentially normal and simply decays in strength inward as the local Mach number drops. But since the penetration region $\delta_0 \leq y \leq y_s$ lies in the outer "velocity-defect" portion of the turbulent profile where the Mach number gradient is very weak, this strength decay is also small except when $y \rightarrow y_s$ (see Fig. 3). For practical high Reynolds number interactions, these facts suggest that a first approximation (and indeed an upper limit) to the shock penetration

effect can be obtained by continuing the full incident shock unchanged across the supersonic region. However, to the same order of approximation, the effect of this is equivalent to a "non-penetrated" interaction solution for an incoming boundary layer having the same skin friction (wall slope) but a smaller thickness $\delta_0' = y_s$. In other words the shock penetration effect in the leading approximation is simply equivalent to a distortion (primarily in thickness) of the boundary layer profile. This idea is easily applied by running the existing program twice: the first run establishes $\delta_0, \eta_s = y_s/\delta_0, C_{f_0}$, etc. and the associated "unpenetrated" interaction field solution, while in the second the programmed equation for δ_0 is multiplied by η_s (as are all values of Re_δ where ever they appear in calculating the Mach number profile, Lighthill sublayer properties, etc.) taking care not to change C_{f_0} ; the results give the interaction field with shock penetration.

DISCUSSION OF RESULTS

Lateral Pressure Gradient Effects

Typical interaction solutions are illustrated in Fig. 4, where both the boundary layer edge and wall pressure distributions are shown. It is seen that the shock-induced lateral pressure gradients are significant within a region of several boundary layer thicknesses upstream and downstream of the shock foot, the wall pressure being higher than the edge ahead of the shock and lower behind it. Further upstream where the wall and edge pressure equalize they decay exponentially with distance. Along the boundary layer edge, a local pressure jump occurs across the shock at $x = 0$ followed by a small region of subsonic post-shock expansion and subsequent recompression, in qualitative agreement with experimental observations^{5, 10}. Far downstream, the wall and edge pressures again equalize and rise monotonically to the final post-shock level like $1/x$.

It is emphasized that the local shock jump at the boundary layer edge and its rapid lateral smoothing across the underlying subsonic flow region that yields a continuous wall pressure distribution are important physical

features that cannot be accounted for without the lateral pressure gradient effect. Indeed, consideration of the former provides one means of checking against other theories and experiment; for example Fig. 5 compares the predicted local interaction-pressure jump as a function of shock strength against a variety of observed values¹¹. Within the validity of the theory (unseparated flow, $M_1 < 1.3$) our results agree well with the data and show the qualitatively-correct trend of approaching the full inviscid Rankine-Hugoniot result with increasing M_1 or Reynolds number.

The inclusion of lateral pressure gradients captures another interesting feature, the existence of a subsonic post-shock expansion region at the boundary layer edge (here due to sign change across the normal shock¹² of the upstream compression waves from the interaction-induced boundary layer thickening, see Fig. 1). Not only is this qualitatively confirmed by experiment and detailed numerical solutions¹³ but also its behaviour with respect to Mach number: as predicted in Fig. 4, Ackeret, Feldman and Rott⁵ have observed that the degree of expansion and the extent of its influence into the boundary layer gradually disappear with increasing Mach number. Moreover, this phenomena plus the aforementioned rapid lateral pressure variations are clearly evident in their detailed pressure surveys of the interaction region (see Fig. 6): outside the boundary layer there is a noticeable overshoot immediately behind the shock of the magnitude shown in Fig. 4.

The influence of Reynolds number on the lateral pressure gradient effect is also significant as shown in Fig. 7. The local shock jump and post-shock expansion increase and decrease, respectively, with increasing Re_L and at sufficiently high values the post-shock expansion region is very small and weak and hence difficult to detect experimentally. An account of this expansion feature is of practical interest not only in understanding the physics of the flow but also in sorting out various theories. For example, mixed transonic flow past a curved wall exhibits post shock pressure dip but for a completely different reason (e.g., as a result of the purely inviscid disturbance flow logarithmic singularity associated with convex surface curvature¹⁴.) To properly differentiate between this effect and other influence requires consideration of $\partial p / \partial y$.

Shock Penetration Effects

The effect on the interaction pressure field as calculated from our approximate model is shown in Fig. 8; the corresponding boundary layer thickening and skin friction are shown in Figs. 9 and 10. It is seen that for the weaker shock cases ($M_1 < 1.15$) the interaction contracts and thins out slightly with increased skin friction, whereas for stronger shocks the reverse becomes true and shock penetration strengthens the interaction by spreading out the streamwise scale, thickening the boundary layer and reducing the local skin friction. The reason for this trend reversal is that shock penetration introduces two effects of opposite sign: for weak shocks the added shock segment acts as an equivalently stronger shock at a slightly higher Mach number, which tends to reduce the spread and strength of the interaction, whereas at higher shock Mach numbers it strengthens the interaction through increased upstream influence and thickening plus hastening of incipient separation, these rapidly becoming the dominant effects.

A parametric study of Reynolds number influence was also carried out; the results are summarized in Figures 11 - 14 showing respectively the relative changes in non-dimensional upstream influence distance (here defined as where $p' = 0.05 \Delta P$), wall pressure, skin friction and displacement thickness growth at the shock foot. These curves further illustrate the intensification of the interaction that occurs with increasing shock strength and also its sensitivity to Reynolds number when $Re_L < 10^8$. For example, while the upstream influence of a $M_1 = 1.20$ shock is increased only 6% by the shock penetration effect at $Re_L = 10^6$, this increase becomes 34% at only a slightly lower value $Re_L = 3 \times 10^6$ (the corresponding thickening under the shock goes from negligible to 40%).

Although the present analysis pertains to transonic unseparated flow, its parametric trends clearly predict that the shock penetration effect increases with both shock strength, Mach number and the onset of separation (equivalent qualitatively to a lower Re_L). This is borne out by Werle and Bertke's finding² (reproduced here in Fig. 15) that omission of this effect in the turbulent case significantly underestimates the strength of the interaction

and its streamwise extent. The present results support their contention that shock penetration is an important feature of interacting high speed turbulent boundary layers. Moreover, to treat it properly the lateral pressure variations must also be included.

REFERENCES

1. Inger, G. R. , "Similitude Properties of High Speed Laminar and Turbulent Boundary Layer Separation", AIAA Paper 76-375, July 1976.
2. Werle, M. J. and S. D. Bertke, "Application of an Interacting Boundary Layer Model to the Supersonic Turbulent Separation Problem", University of Cincinnati Dept. of Aerospace Eng. Report AFL 76-4-21, Aug. 1976.
3. Shang, J.S. Hankey, W.L., Jr., "Numerical Solution for Supersonic Turbulent Flow Over a Compression Ramp", AIAA Journal, Vol. 13, pp. 1368-74, October 1975
4. Inger, G. R. and W. H. Mason, "Analytical Theory of Transonic Normal Shock-Turbulent-Boundary Layer Interaction", AIAA Journal 14, Sept. 1976, pp. 1266 - 72.
5. Ackeret, J., F. Feldman, and N. Rott, "Investigations of Compression Shocks and Boundary Layers in Gases Moving at High Speed", NACA TM-1113, Jan. 1947.
6. Inger, G. R. , "Analysis of Transonic Normal Shock-Boundary Layer Interaction and Comparisons with Experiment," AIAA Paper 76-331 (9th Fluid and Plasma Dyn. Meeting, San Diego), July 1976, Also VPI SU Report Aero-053 , Blacksburg.
7. Inger, G. R. , "Analysis of Transonic Shock Interaction with Nonadi Turbulent Boundary Layers," AIAA Paper 76-463 (11th Thermophysics Conf. , San Diego), July 1976.

8. Lighthill, M. J. , "On Boundary Layers and Upstream Influence; II, Supersonic Flow Without Separation," Proc. Royal Soc. , A217, No. 1131, pp. 478-507, 1953.
9. Mason, W. M. and G. R. Inger, "Analytical Study Transonic Normal Shock-Boundary Layer Interaction," VPI &SU Report Aero-027, Blackburg, Nov. 1974
10. Gadd, G. E. , "Interactions between Normal Shock Waves and Turbulent Boundary-Layers," British ARC Ref. 22559 (N. P. L. R & M 3262), Feb. 1961
11. Lomax, M. , F. R. Bailey, and W. F. Ballhaus, "On the Numerical Simulation Three-Dimensional Transonic Flow with Application to the C-141 Wing," NASA TN D-6933, Aug. 1973.
12. Adams, M. C. , " On Shock Waves in Inhomogeneous Flow," Jour. of the Aero.Sci. Vol. 16, No. 9, pp. 685-590; September 1949
13. Melnick, R. and M Grossman, "Analysis of the Interaction of a Weak Normal Shock Wave with a Turbulent Boundary Layer," AIAA Paper 74-598, June 1974.
14. Oswatitsch, K. and J. Zierep, "Das Problem des Senkrechten Stoßes an einer gekrümmten Wand," ZAMM, Bd. 40 (1960), p. 143.

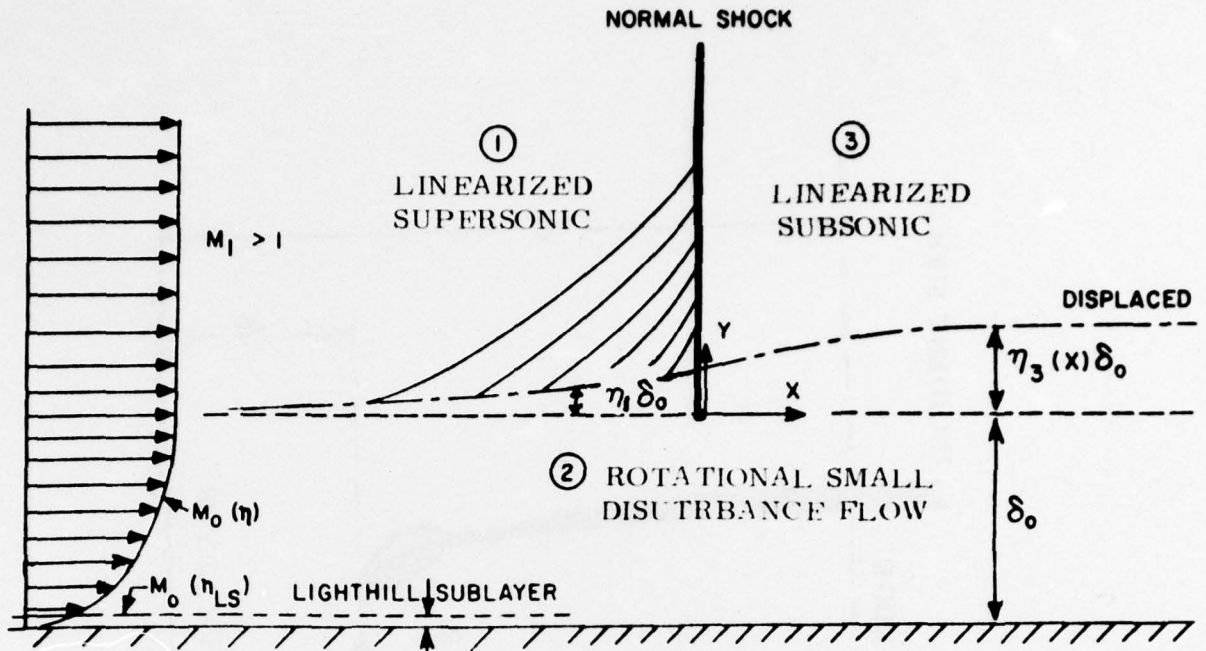


Fig. 1. Basic Interaction flow Model

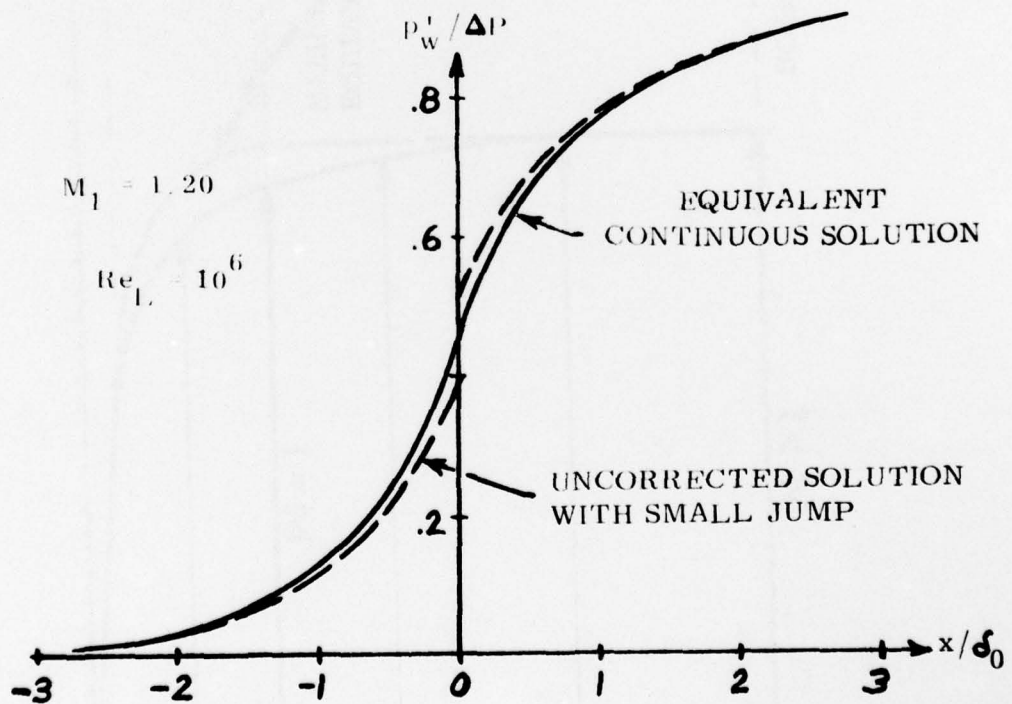


Fig. 2 Wall Pressure Solution Neglecting Shock Penetration Effect

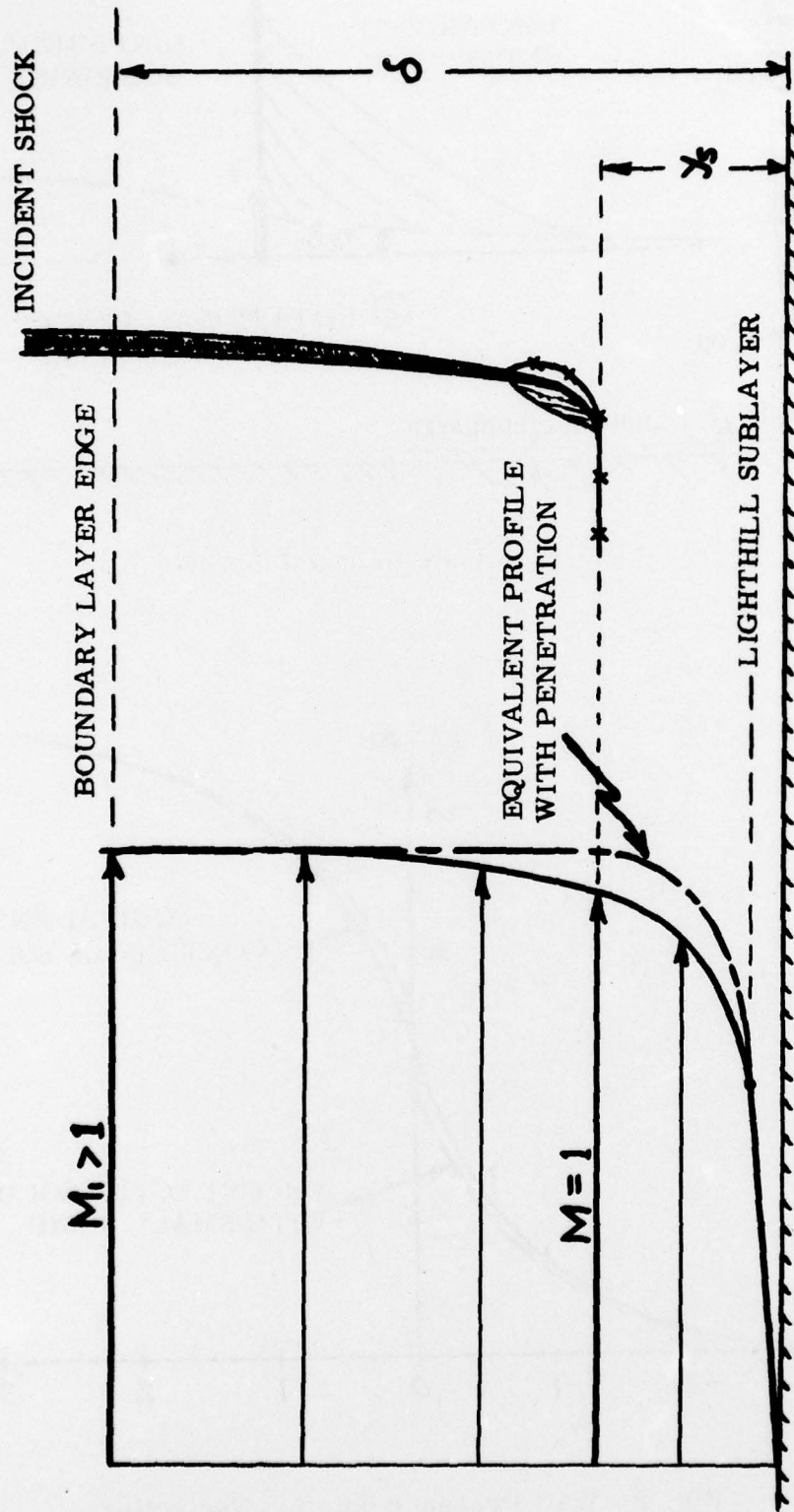


Fig. 3 Shock Penetration Region (Schematic)

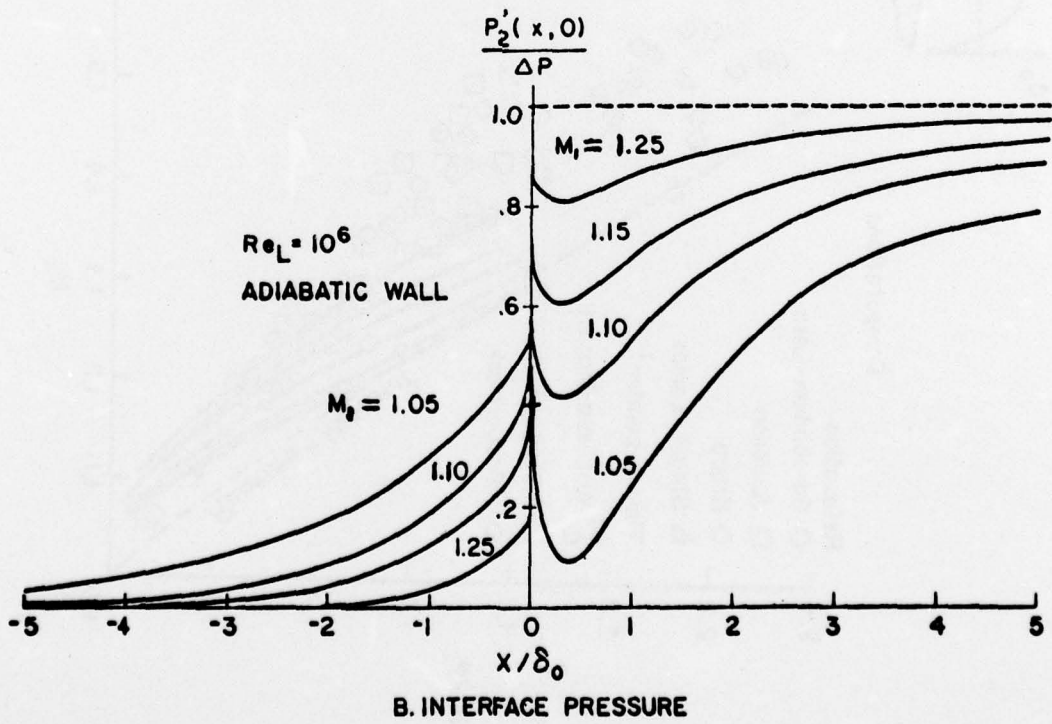
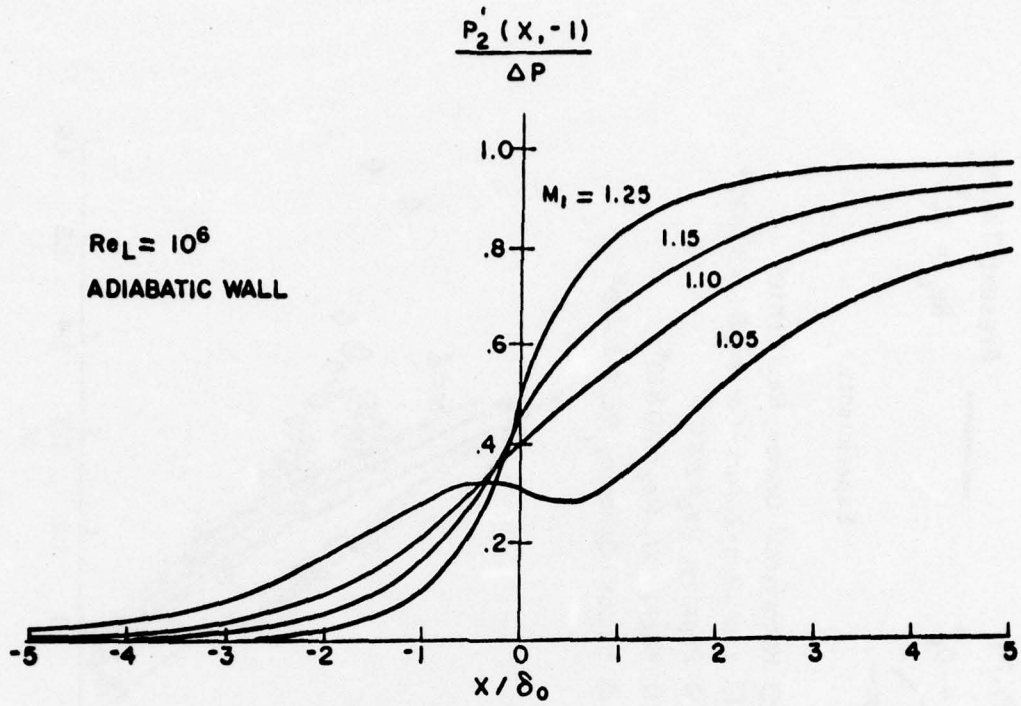


Fig. 4 Influence of Shock Strength on Interaction Pressure Field

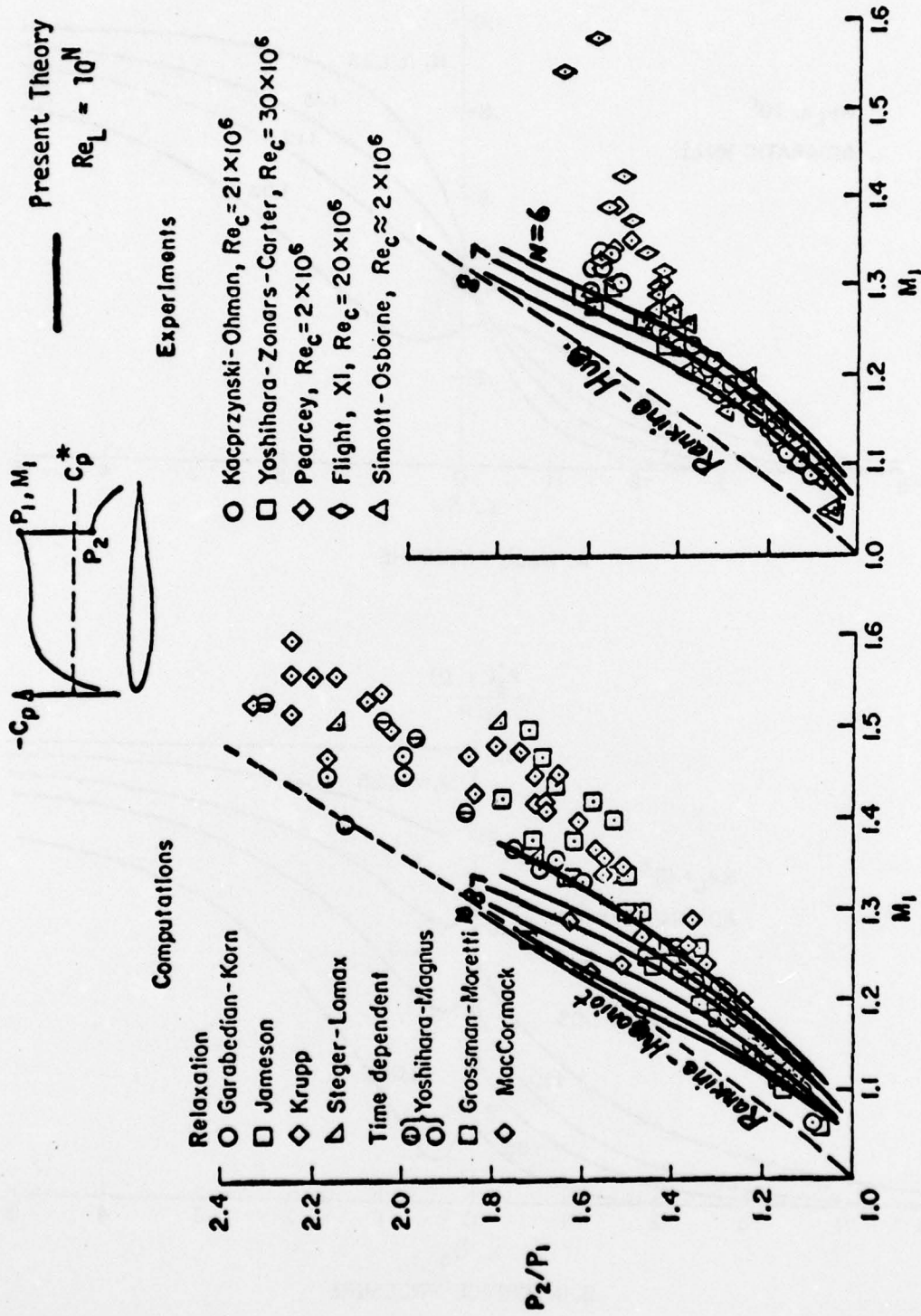


Fig. 5 Comparison of Theory and Experiment for Local Shock Wave Pressure Jump

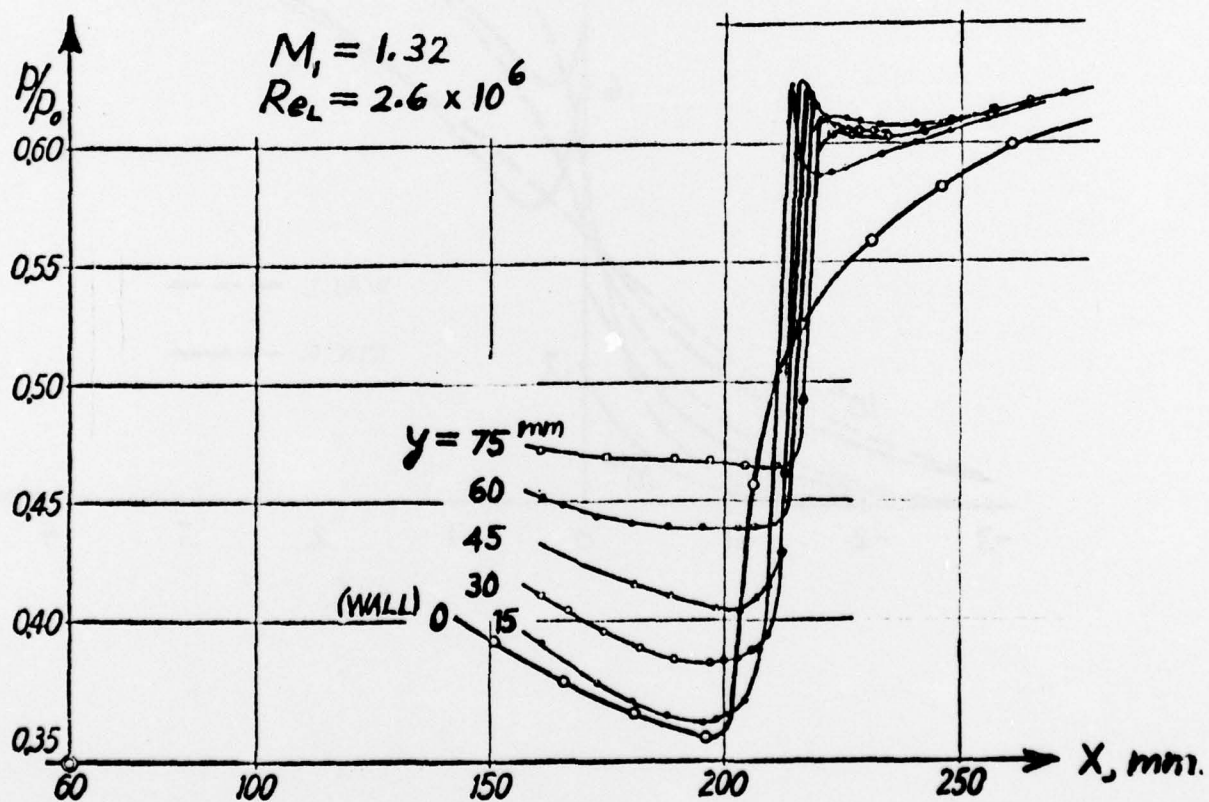
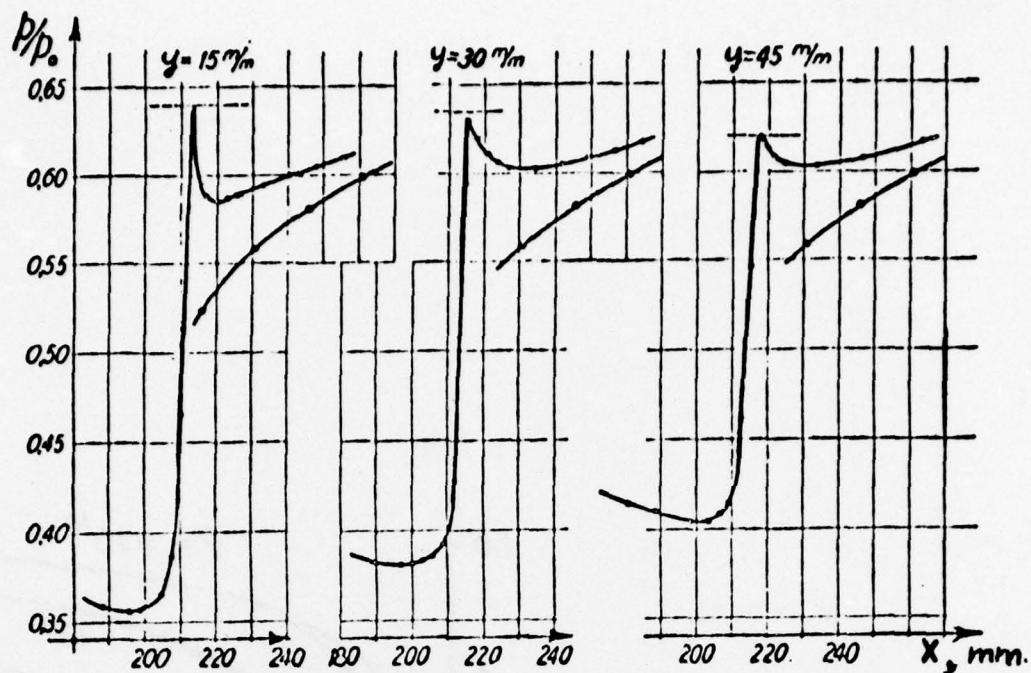


Fig. 6 Typical Experimentally Observed Lateral Static Pressure Variations Across Interaction Region⁵

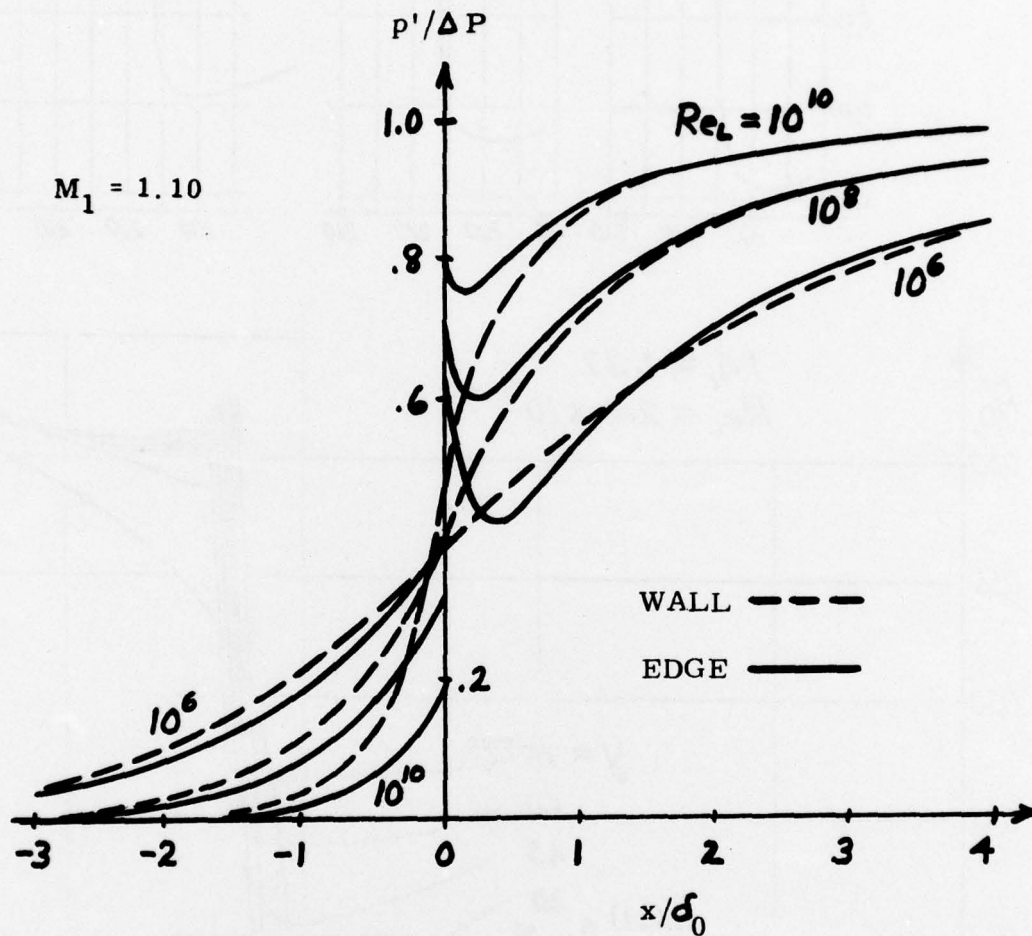


Fig. 7 Reynolds No. Effect on Interaction
Pressure Field Across Boundary Layer

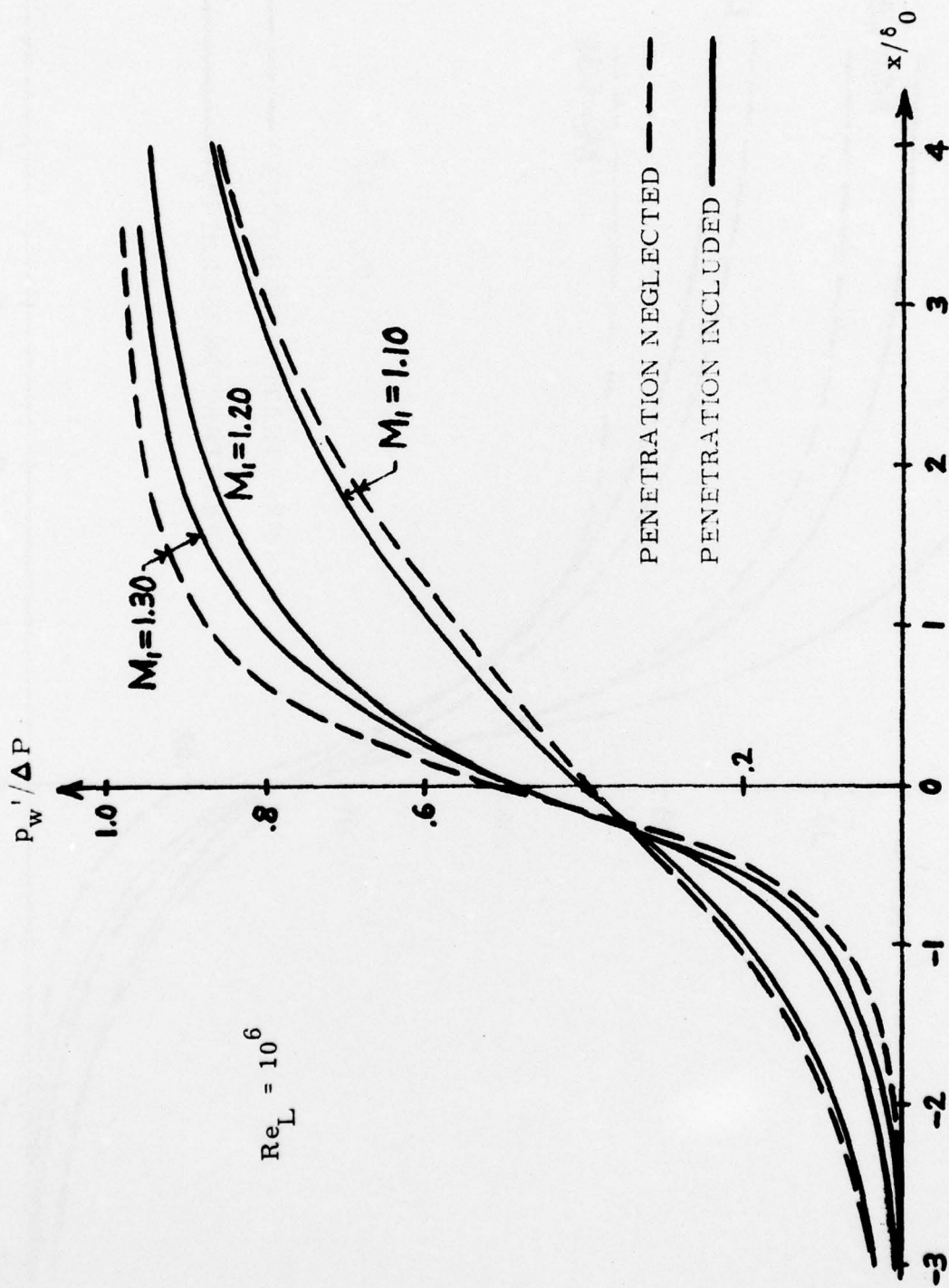


Fig. 8 Shock Penetration Effect on Wall Pressure

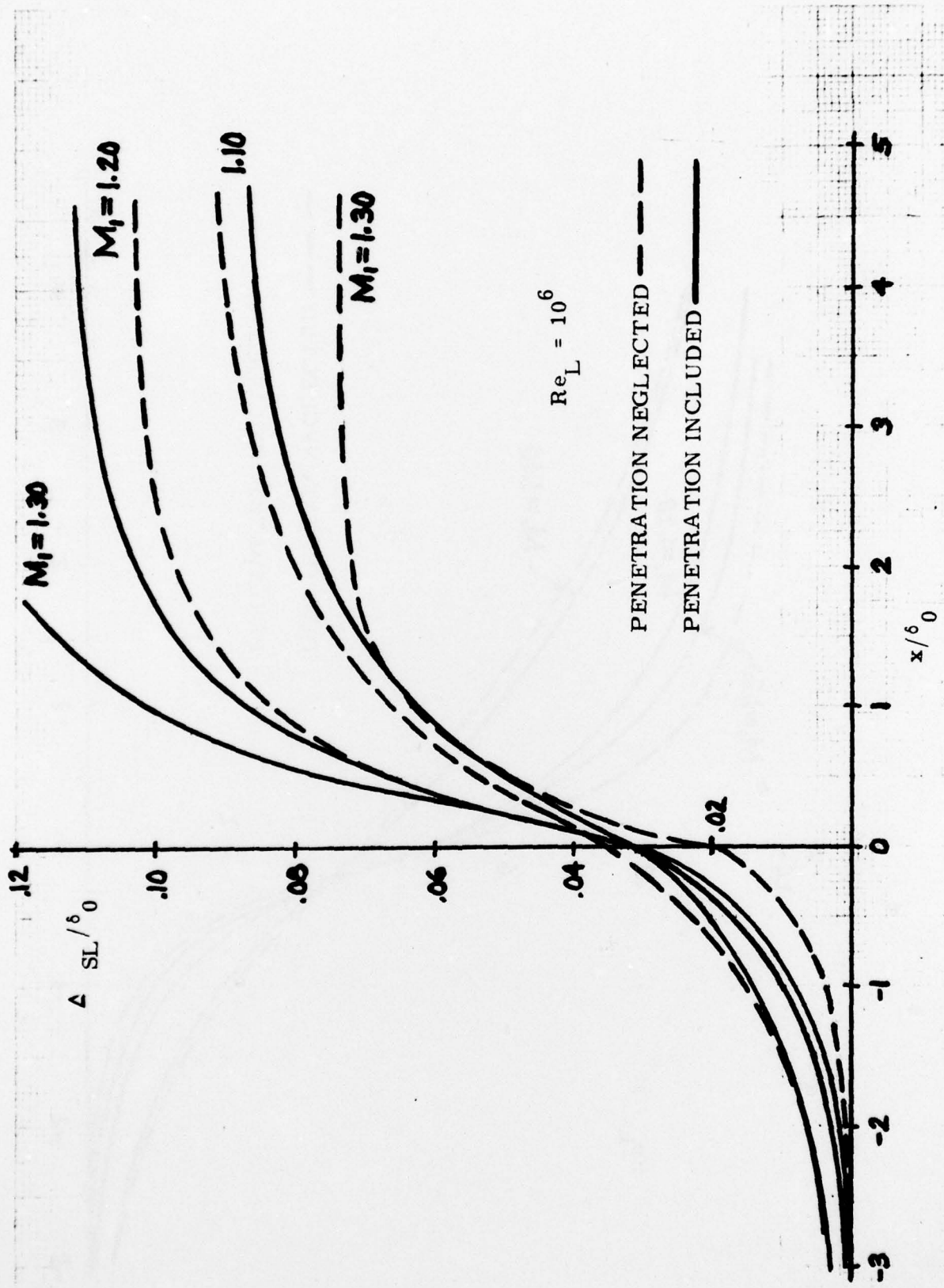


Fig. 9 Shock Penetration Effect on Interaction Thickening

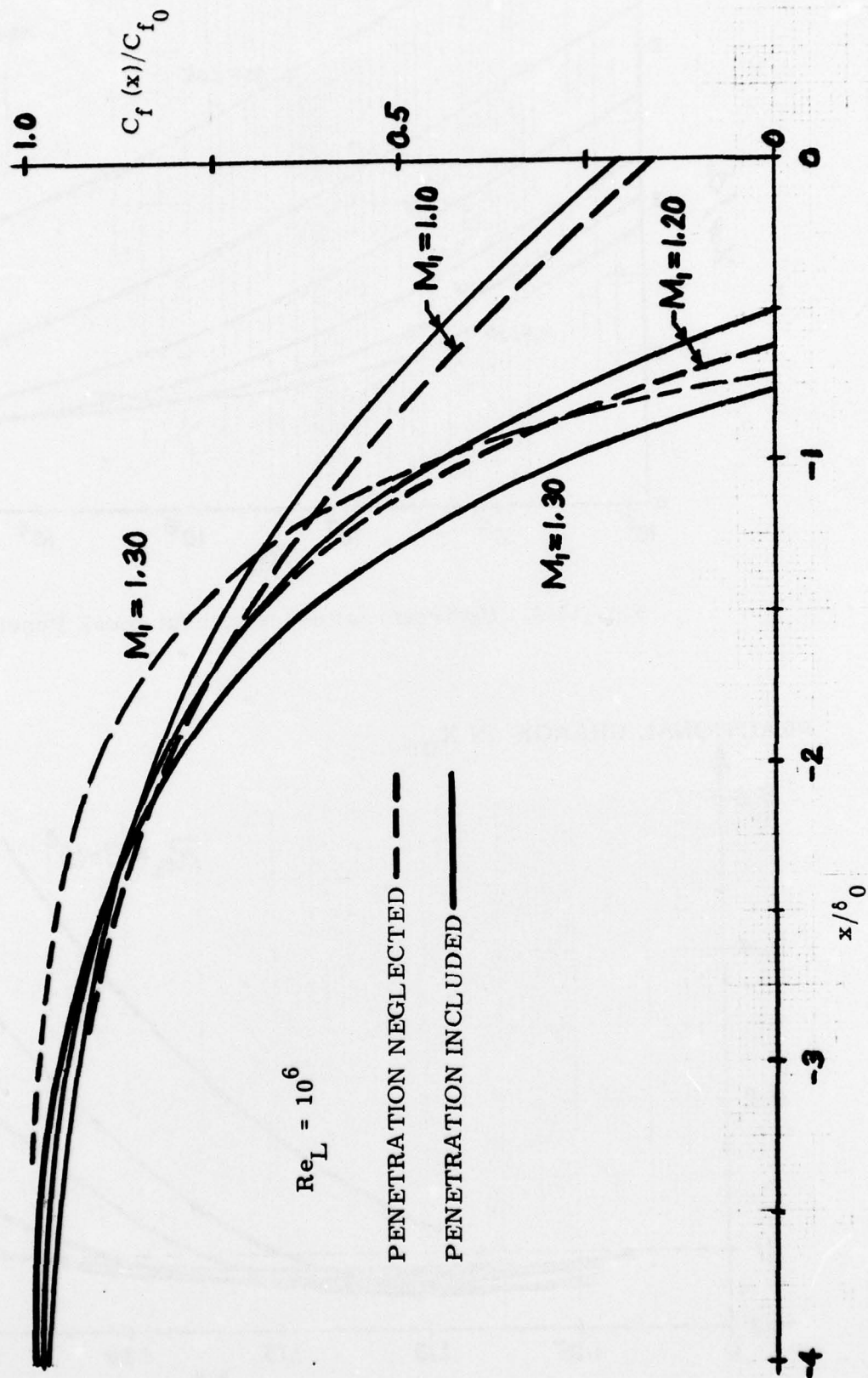


Fig. 10 Shock Penetration Effect on Skin Friction

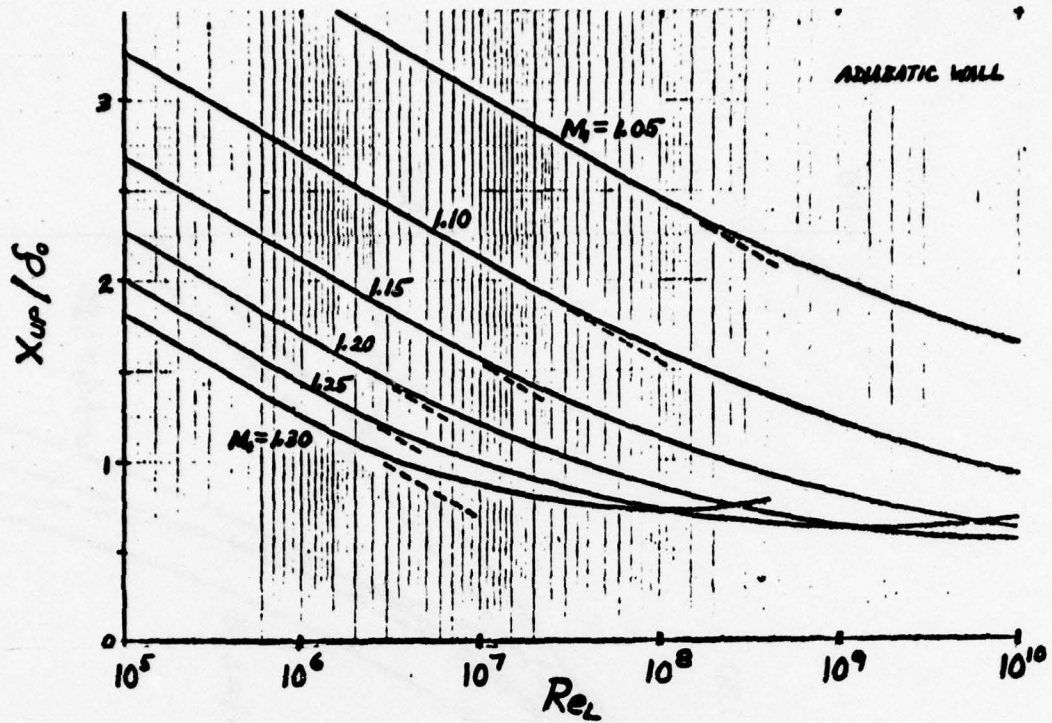


Fig. 11 A. Upstream Influence without shock Penetration

FRACTIONAL CHANGE IN X_{UP}

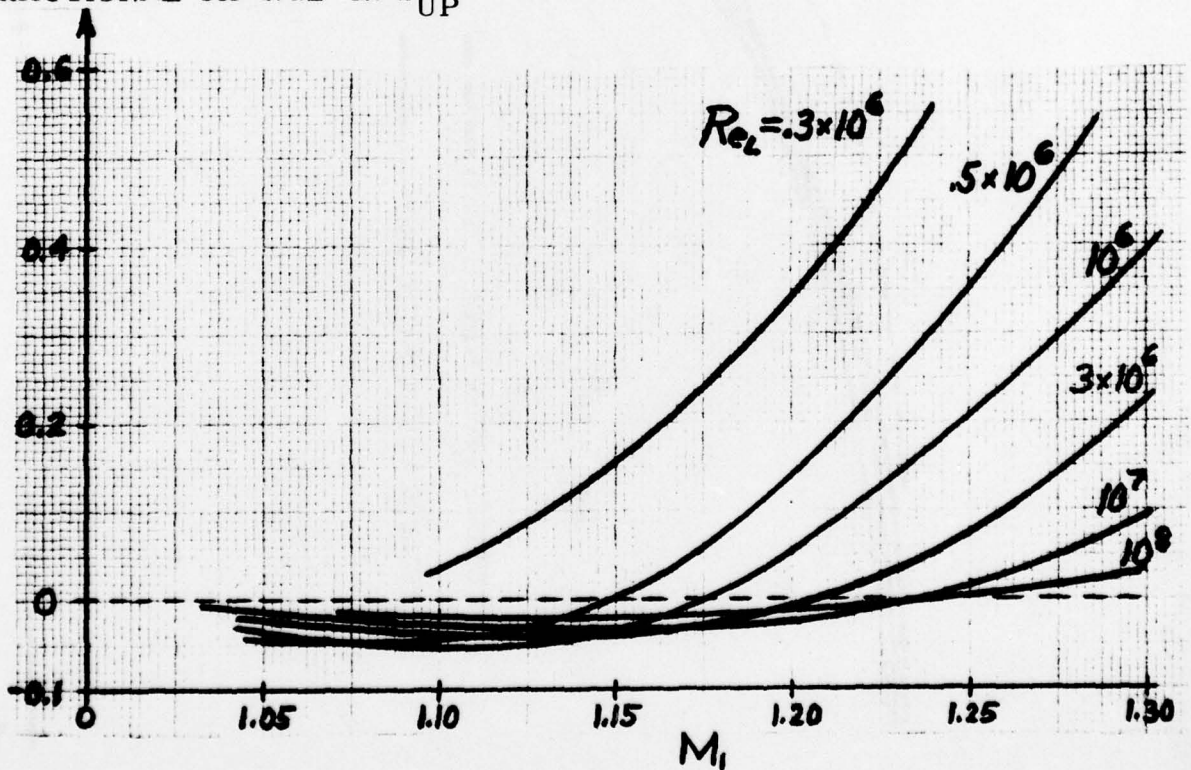


Fig. 11B Shock Penetration Effect on Upstream Influence

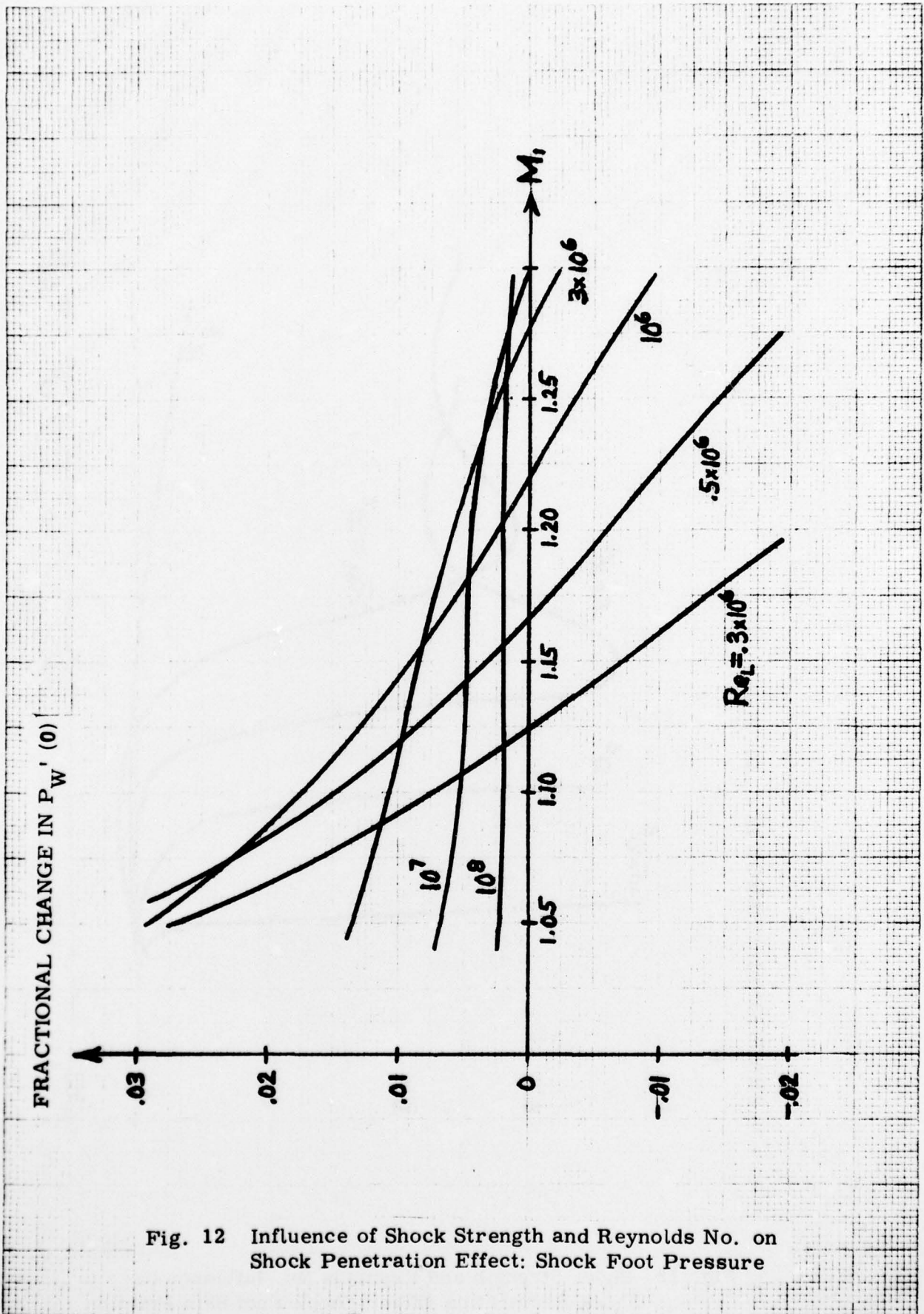


Fig. 12 Influence of Shock Strength and Reynolds No. on Shock Penetration Effect: Shock Foot Pressure

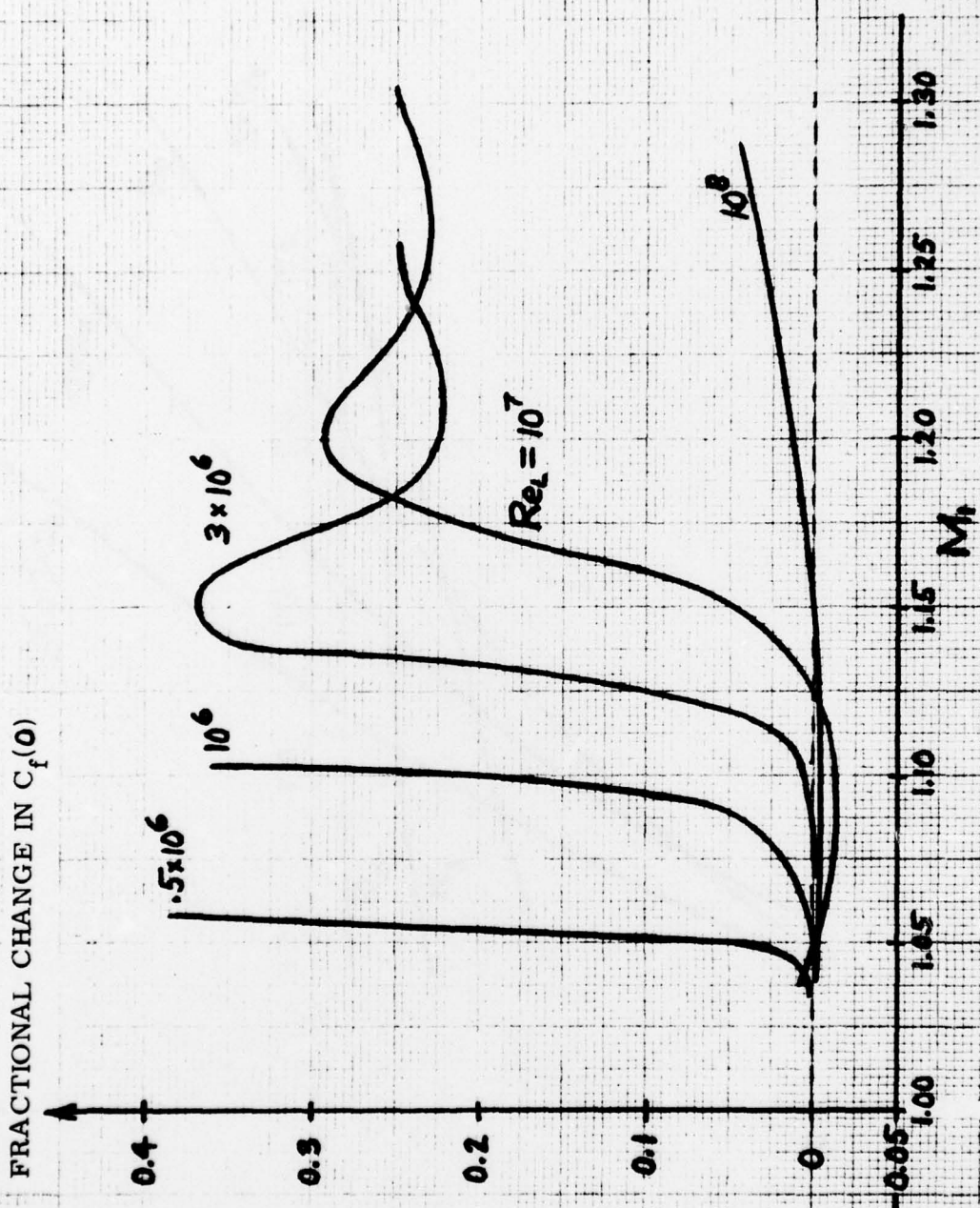


Fig. 13 Shock Strength and Reynolds No. Influence on Shock Penetration Effect: Shock Foot Skin Friction

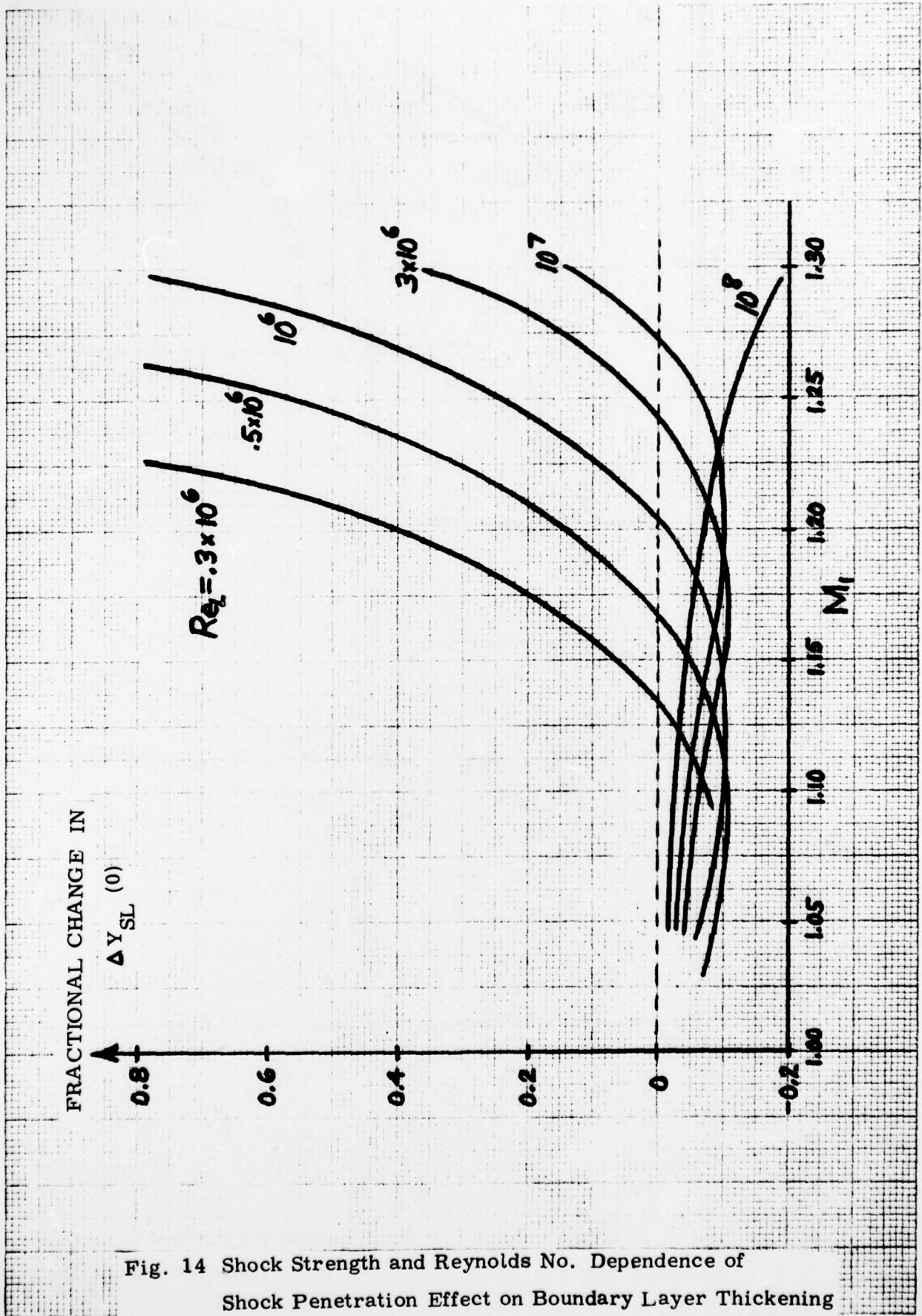


Fig. 14 Shock Strength and Reynolds No. Dependence of Shock Penetration Effect on Boundary Layer Thickening

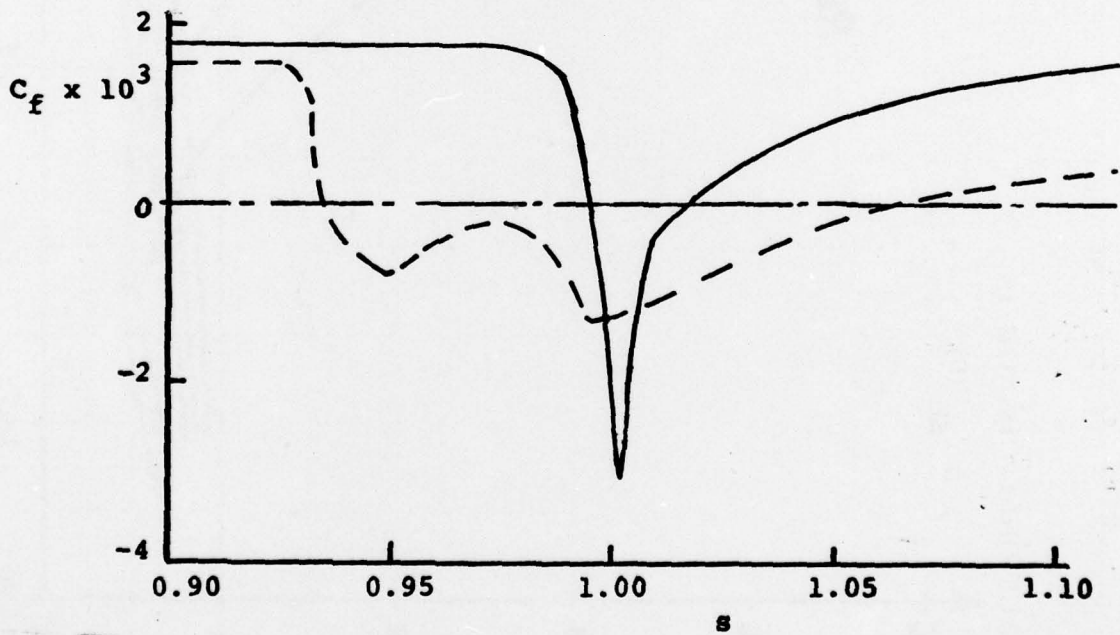
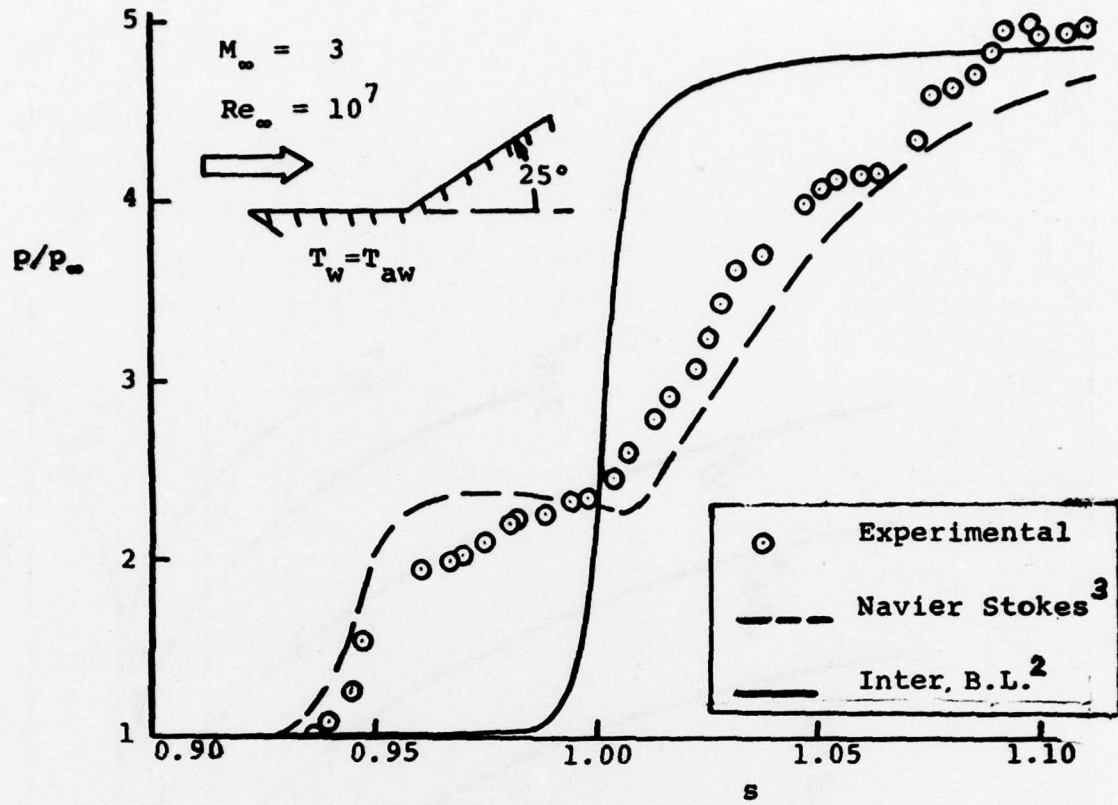


Fig. 15

Comparison of Theories and Experiments for Surface Pressure and Skin Friction on ARL Ramp (from Werle and Bertke)

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