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INVESTIGATION OF FUNCTIONAL COMMONALITY OF

AVIONICS SYSTEMS IN NAVAL AIRCRAFT

30 SEPTEMBER 1981

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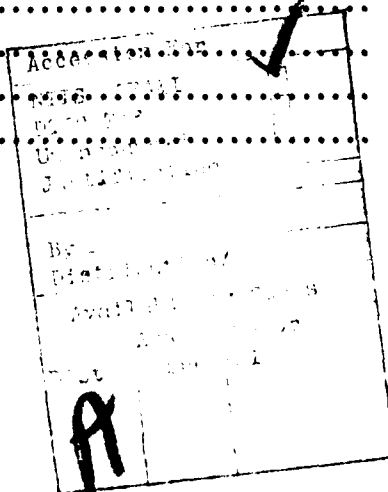


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1.0 INTRODUCTION

This document represents the results of a study commissioned under AIR-TASK A533533F/001D/1W05720000 in support of the NAVAIR Avionics Components and Subsystems Program. The background and objectives of the overall program as stated in reference (e) are:

Background

"A major concern within the military avionics community is the proliferation of unique avionics equipment that increases with each new aircraft development. Because of the limited quantities of avionics equipment associated with any particular aircraft, a growing cost burden has been experienced in the areas of multiple subsystem development, procurement, logistics, maintenance, and total subsystem life cycle cost (LCC). The Avionics Components and Subsystems Program (AVCS) program was formulated by NAVAIR to reduce proliferation of unique avionics equipment and life cycle costs by providing a family of Government Furnished Equipment (GFE) common to multiple aircraft. The AVCS program provides for the development of avionics equipment supportive of, but separate from, major weapon system acquisitions."

Objectives

"The objectives of the AVCS program are listed below:

- (1) provide comprehensive planning and analysis of current and future user needs,
- (2) reduce proliferation and associated costs of unique weapon system avionics,

- (3) maximize commonality of hardware and software across multiple aircraft,
- (4) maximize flexibility and growth potential by careful subsystem architecture design and incorporation of digital data bus interface capability,
- (5) select mature technologies, suitable for military avionics applications, that will be available and logistically supportable for the projected service life of the equipment,
- (6) minimize the need for organizational level Ground Support Equipment (GSE) by careful system architecture design and functional partitioning, and by implementing Built-In-Test (BIT) and In-Flight Performance Monitoring (IFPM) to meet fault detection and isolation requirements, and
- (7) provide for the development of common GFE avionics, separate from major weapon system acquisition, that will receive rigorous testing to assure reliability and supportability."

This report was prepared in direct support of objective (1) in order to provide recommendations relative to achieving the remaining AVCS program objectives.

The basic objective of the AVCS program is commonality among avionics in different platforms. The few elements of Naval avionics that are common today primarily came about because of GFE (which in many cases was developed as contractor furnished equipment (CFE)). From the weapons system point of view, GFE is equipment that must be adapted to make it work since the GFE is rarely directly applicable to a new or upgraded weapons system architecture. The basic problem is that the weapons system contractor is provided a piece of

equipment (GFE) which was not designed as a part of the weapons system in which it is being installed. When an upgrade of a subsystem or element is necessary, it is often made as a "form, fit, functional replacement." What occurs in reality is that boxes are built to be compatible with the adaptations made in the weapons system to accommodate the original GFE. For the GFE equipment items of the AVCS program to be completely successful, they must be designed as part of a weapons system and not as pieces of equipment.

It is not probable that the Navy will ever have an ideal weapons system in which every element is designed to do exactly what is necessary to support the mission of that weapons system. If technological superiority is to be maintained within the limits of aircraft constraints, then an attempt must be made to approach the ideal weapons system. This attempt is most likely to require higher levels of subsystem integration in order to achieve increased capabilities with finite resources. In order to make AVCS compatible with future avionics needs common equipment must be integratable. The primary goal of this study is to define ways of insuring that future avionics equipment items are elements of a weapons system that can be used across a range of weapon systems and are compatible with future growth (increased integration).

2.0 APPROACH

In the preparation of this report, the analysis of current and future needs for weapons system avionics, with particular emphasis on multiplatform avionics requirements, was attempted. The analysis followed this path in inquiry:

- o What does the Navy need in terms of a modern, flexible, practical avionics suite (overall and core avionics requirements)?
- o What architecture can be used to satisfy the need (architecture design)?

- o What are the specific subsystem requirements of the architecture (interfaces to the architecture)? and
- o What action is necessary to allow the architecture to evolve (recommendations)?

The effectivity of the results of this process would appear to be highly dependent on the practical acceptance of the architectural design. For some of the recommended hardware developments, this is true. The recommendations regarding interface requirements, standards, and procurement practices, however, were made as independent of the specific architecture as possible. This was done by attempting to address the interfacing of equipment from a functional point of view.

3.0 GENERAL ARCHITECTURAL REQUIREMENTS

The purpose of this report is to develop a multiplatform avionics architectural concept that allows emerging technologies and the realities of current and future weapon systems to come together. The requirements for a common architecture that would ideally satisfy the Navy's needs are:

- a.) Technology transparent functional partitioning
- b.) Compatibility with the accommodation of specialized subsystems
- c.) Survivability
- d.) Maximum commonality
- e.) Efficient hardware utilization
- f.) Minimization of interconnect requirements
- g.) Elimination of superfluous redundancy
- h.) Sufficient throughput capability
- i.) Applicability to multiple platforms
- j.) Compatibility with Red/Black requirements
- k.) Provision of simple control and display

- l.) Compatibility with procurement
- m.) Allowance for growth.

3.1 ASSESSMENT OF REQUIREMENT

The ideal architecture requirements cannot be met in any real sense. It is important to realize that many of the requirements are diametrically opposed and, as a result, any architecture will trade-off the requirements against each other. One method of resolving the conflicts between requirements is to prioritize them. However, rigidly enforced priorities cannot allow for intelligent resolution of specific conflicts. The best way to achieve an optimum architecture is to examine all the conflicts and make decisions on a problem by problem basis. In the following discussion each item in paragraph 3.0 is discussed and the conflicts with other items in paragraph 3.0 are then examined.*

- a.) Technology Transparent Functional Partitioning. This is one area in which very few conflicts exist since it is a natural fallout of a properly organized architecture.

- ③ Efficient hardware utilization in some cases would require the removal of functional partitions; however, in order to maintain growth capability and flexibility, the logical partitioning of functions, particularly to allow technology transparency, should not be redirected solely to achieve efficient hardware utilization.

* The circled letters refer to items in paragraph 3.0 which are in conflict with the item under discussion.

- b.) Compatibility with Accommodating Specialized Subsystems. This requirement creates the greatest number of conflicts since it explicitly states that the architecture must accommodate, but not include, certain functions.
- (c) The departure from any architecture has survivability impact, particularly in accommodating fault tolerance and redundancy requirements. Provision within the architecture for survivability of unforeseeable requirements is not considered prudent. Hence, survivability issues for specialized subsystems are part of that subsystem and affect the architecture only where interfaces with the architecture are required.
- (d) Maximum commonality cannot be achieved when specialized subsystems are allowed. Enforcing commonality on a specialized subsystem is usually unsuccessful and often inefficient. If sufficient standards are applied then interfaces, whether or not they connect to the architecture, can and should be made compatible. Common modules should be required only when they do not impact overall subsystem design.
- (e) Efficient hardware utilization does not allow specialized subsystems. This conflict should be resolved when the decision is made to allow a subsystem to be specialized. A specialized subsystem is required when the basic architecture cannot do the job; however, there are in all probability one or more parts of the job that can be done within the architecture. Hence, when a subsystem is declared special, those functions of that subsystem that are to be performed within the architecture should be explicitly stated to allow maximum hardware utilization.
- (f) Minimizing interconnect requirements cannot be accomplished effectively with specialized subsystems, and only cursory requirements should be applied in this area.

- ⑧ Superfluous redundancy will result in a specialized subsystem when unnecessary redundancy for the subsystem is implemented. Eliminating the superfluous redundancy by utilizing the architecture is rarely a possibility because, by definition, if the architecture could do the job, the subsystem would not need to be specialized. If the mission critical portion of the subsystem can be accomplished within the architecture, then the specialized subsystem is being utilized only for accessory functions. Superfluous redundancy can be eliminated for specialized subsystems by limiting the redundancy requirements and following the guidelines above for efficient hardware utilization.
- c.) Survivability. The survivability requirements fall into two areas, namely, redundancy and fault tolerance. In addition, survivability should be looked at from two separate perspectives: having a high probability of maintaining mission critical capabilities with limited weapon system damage (this is often carried to extremes where non-flight safety capabilities are attempted to be maintained in the presence of damage that would not let the aircraft fly), and maintaining safety of flight critical functions to the maximum extent possible.
- ⑨ Efficient hardware utilization is not in conflict with fault tolerance but is opposed to redundancy. Efficient hardware utilization can be achieved if redundancy is required only where it is necessary. Redundancy for safety of flight critical functions is considered an absolute requirement. Redundancy for any other purpose should not be incorporated unless justified in detail.
- ⑩ Minimizing interconnect requirements is not in conflict with redundancy once redundancy is required. Fault tolerance and minimizing interconnects are, however, in conflict. Fault tol-

erance cannot make a fault go away; redundancy does that. The impact of faults can be minimized by partial replacement of functions by other assets (a form of redundancy), but normally this requires additional interconnects to facilitate the assumption of tasks. Additional interconnects to facilitate partial replacements of functions should only be incorporated for mission critical capabilities and only when the partial function replacement allows performance of the mission critical capability.

- (k) Simple control and display does not lend itself to either redundancy or fault tolerance. Loss of prime capability and reduced capability affected by fault tolerance features must be reported but can result in cluttered displays. Requirements for redundancy in the displays and controls further complicate the problem of simplifying the crew interfaces. Multiple displays and controls for the same purpose are at times confusing. Limitations on redundancy requirements can minimize the problem but the problem still exists. Use of degraded mode backup as opposed to redundant displays and controls (i.e., maintenance of critical capability via deletion of a noncritical capability) is a useful solution as long as multifunction devices are utilized and they appear the same for the critical functions in the degraded mode. Use of compatible, if not identical, displays and controls for all crew stations allows transfer of an entire function to another crew member and is an effective means of obtaining redundant capability. The requirements for true redundant system display and control can only be intelligently resolved on a control by control, display by display basis. Fault tolerance display and control issues also require individual resolution.

d.) Maximum Commonality is desirable in any architecture because it simplifies the constraints on the architecture but it has impact in several areas.

- ⑧ Sufficient throughput capability for the worst case application leads to excess capability for the remainder of applications when common elements among platforms are used. Whenever possible, peaks in throughput requirements should be handled by parallel capabilities of common elements. This approach leads to some inefficiencies, but should be utilized to the maximum extent possible because it has inherent growth capability.
- ① Multiple platform applications necessitate satisfying peculiar requirements or there would be no need for multiple platforms. Significant potential exists for maximizing commonality in newly developed platforms. An architecture that capitalizes on commonality is difficult to incorporate in existing platforms unless the existing hardware is used as the common element. Accommodating existing platforms is perhaps best handled by modularizing all input/output (I/O) elements within the architecture and allowing for variants for existing platforms.
- ① The procurement of subsystems is severely complicated by commonality with other subsystem constraints. Procurements can state "how" or "what." Typically, they say "what," at least for the initial procurement. Enforcing commonality requires some "how" along with the "what" in procurement. GFE of common elements is possible but complicates procurement and accountability of the contractor. In addition, the elements must exist for GFE which requires serial rather than parallel procurements.

e.) Efficient Hardware Utilization implies that if one piece of hardware can perform the same function for two elements of the weapon system, then it is assigned the tasks.

⑥ The equipment interfaces necessary for efficient hardware utilization often increase the interconnect requirements to accommodate the sharing of resources. In general, the highest cost benefit is achieved by maximizing the hardware utilization.

⑦ Red/Black requirements, in some cases, prevent one piece of hardware from doing two things, but the requirements must be adhered to. However, the architecture should arrange Red/Black signal distribution to allow maximum hardware utilization.

f.) Minimizes Interconnect Requirements. Minimizing interconnects saves cost, weight, and maintenance, and increases survivability and reliability.

⑧ Allowances for growth involve providing for the unknown. If interfaces are minimized, the probability that the interface that is required for growth exists is diminished. One approach that may actually minimize interconnects is to collect all information at a central location. This approach, works only when the timing of the information is not critical, and can have a severe influence on throughput and survivability. There is no optimal solution for resolving the conflict between growth and interconnects, but standardized general purpose inputs and outputs offer the most growth potential.

g.) No Superfluous Redundancy. Excessive redundancy is very attractive to architectural design, since it is compatible with growth and provides sufficient throughput (i.e., the excess can be used later or

in special cases since there is no real requirement for redundancy). Unfortunately, once something is labeled redundant, it is very difficult to remove the label to capitalize on the capability. If redundancy is not required, it should not be added.

- h.) Sufficient Throughput Capability is self-explanatory, except in the definition of sufficient. No valid architecture should ever reach the maximum throughput limit, but how much excess should there be? An arbitrary answer is 10% for the known worst case requirement when all potential platforms are considered.
- i.) Applicability to Multiple Platforms. For a generalized architectural concept to be of value, it must be applicable to at least two and preferably all platforms.
- j.) Compatibility with Red/Black Requirements is not considered a negotiable area.
- k.) Simplified Control and Display is a very difficult area in which to achieve a self-explanatory requirement. However, if the control and display is not sufficiently clear, the capabilities designed into the architecture will probably never be used.
- l.) Compatible with Procurement. If the architecture does not allow for subsystem performance specifications to be written and tested, then the architecture cannot be implemented. This is a major consideration that must be included in the architecture.
- m.) Allows Growth. Compliance with this requirement should be met in the sense of growth with no modification to existing elements whenever possible.

3.2 RESULTS OF CONFLICTING REQUIREMENTS ANALYSIS

The conflict resolutions discussed above did not resolve the conflicts at all, but they did constrain some of the architectural considerations. More questions were asked than were answered and more "don'ts" than "do's" were generated, but some guidelines are clear, viz.:

- o Specialized subsystem survivability considered only at interface
- o Commonality only among architectural members
- o Specialized subsystems should have some functions done for them within the architecture
- o Require redundancy only when actually necessary
- o Additional interconnects for fault tolerance only for mission critical functions
- o Modularized I/O can help incorporation in existing platforms
- o Throughput should be expandable in a parallel fashion
- o Red/Black requirements should not be compromised
- o Use general purpose interconnects to the maximum extent possible
- o Maintain at least 10% reserve throughput capability.

4.0 ARCHITECTURAL DESIGN

The design of an aircraft avionics architecture is highly dependent on the mission requirements of the weapon system. This is true primarily because of the level of subsystem integration necessary to achieve overall weapon system capability within size, weight and cost constraints. In modern Naval aircraft the extent of subsystem integration required has, for all practical purposes, obliterated the meaning of the word system. A function is performed by a group of subsystems none of which exist solely for the performance of that function, e.g., a bombing system is made up of an ordnance subsystem, a radar subsystem, a display and control subsystem, a navigation subsystem, a communications subsystem, and the flight control subsystem, all of which are used to

perform other functions (make up other systems). The mix of equipment types and the integration requirements vary by aircraft mission, however, certain functional elements are required for all aircraft. The basic requirements for the functional elements that are common between aircraft (core avionics) differ only in the integration requirements.

Partially to accommodate the integration requirements of different aircraft a variety of similar but different core avionics equipment that have the same basic functional requirements have been developed. By considering the realities of the systems integration requirements in the design of core avionics equipment, functional commonality between aircraft types can be achieved without limiting the weapon system avionics architecture design.

In order to investigate the integration requirements of avionics equipment, a multiplatform avionics architecture that addresses only core avionics equipment was generated as part of this study. The architecture is not intended as a blueprint for future aircraft but it does serve several purposes:

- a. The model generated attempts to demonstrate a method of incorporating the general architectural requirements discussed in Section 3.0 while maintaining the capability of accommodating the variable integration requirements of each of the platforms.
- b. The architecture attempts to provide a mechanism whereby existing equipment can be accommodated in future weapon systems.
- c. The architecture separates performance requirements of individual subsystems to the maximum extent practical in order to show how the orderly incorporation of improvements which are compatible with procurement and test requirements can be accomplished.

- d. The architecture demonstrates a means of accommodating the transplantation of new developments in integrated subsystems.
- e. In addition, the architecture highlights those areas where, for reliability reasons or because of additional requirements on the subsystem, the subsystem integration efforts will require some variations between platforms.

4.1 CORE AVIONICS SUBSYSTEMS

The subsystems considered as core avionics are:

- o Flight Control
 - Dynamics Sensors
 - ACLS (Automatic Carrier Landing System)
- o Flight Instrumentation
 - Velocity
 - Attitude
 - Altitude
 - Heading
- o Navigation
 - Inertial Navigation
 - TACAN (Tactical Air Navigation)
 - ADF (Automatic Direction Finding)
- o Communications
 - VHF/UHF (Very High Frequency/Ultra High Frequency) Voice
 - UHF Data

- ICS (Intercommunications System)
- IFF (Identification Friend or Foe)
- o Displays and Controls

There are a number of subsystems that were excluded from this list because of a lack of universal applicability (e.g., HF (High Frequency) communications, LORAN (Long Range Aid to Navigation), Doppler Navigation). There are also several subsystems that are not universally applicable (e.g., the Radar Beacon and the ILS (Instrument Landing System) components of ACLS) that were included because of universal applicability to CV (Aircraft Carrier) operations.

4.2 SAFETY OF FLIGHT REQUIREMENTS

The Mission Essential Subsystem Matrices (MESM's) of reference k were examined for the A-6E, F-14A, F/TF/A-18, and P-3C weapon systems. For the core avionics subsystems in 4.1 the following requirements for safety of flight were determined.

- o Redundancy for:
 - Flight Control System Components
 - Flight Instrumentation
 - Intercommunications
- o Backup capability
 - Redundant UHF or VHF backup for UHF
 - TACAN with ADF as backup

4.3 MULTIPLATFORM AVIONICS ARCHITECTURE

The multiplatform avionics architecture developed for the core avionics subsystems is shown in Figure 1 (provided as a foldout at the end of this report). The architecture is to a large extent based on utilizing existing

equipment types. The Control and Display Subsystem is based on the Advanced Integrated Display System (AIDS) and the navigation and flight control sensors are combined using an Integrated Inertial Sensor Assembly (IISA).

The architecture is applicable to the near term procurement of a new weapon system which is the main reason for reliance on existing equipment types. The use of existing equipment types results in substantial similarities between Figure 1 and the actual implementations in the F-14A and F/A-18. The rationale for using AIDS and IISA concepts for a near term procurement is that both the display and control and overall flight control subsystems are assumed to require development for any new weapon system.

The central element in the architecture is the Mission Computer. Redundancy for the Mission Computer has been assumed, however, the actual number of computers, the functional allocations between the computers, and the computer architecture will vary by aircraft. The Mission Computer(s) are assumed to be CFE equipment (even though hardware and some software modules may be government furnished). The Mission Computer(s) are considered CFE because of the differences in functions and software required for various platforms. Because the Mission Computer(s) will require some CFE software, they cannot be supplied as an entire functioning unit and are therefore not labeled as CFE. The Mission Computer(s) is responsible for all core avionics management functions (except flight control) and as such forms a central point to which all other equipment must be interfaced. The form of interface with the Mission Computers has been assumed to be via time multiplexing of digital information. Although for clarity not illustrated in Figure 1, all time multiplex interfaces in Figure 1 are assumed dual redundant. Each of the interfaces represent composite signal flow paths and not necessarily individual wires.

A preflight memory device is recommended to load mission variables and miscellaneous information for all subsystems. This device would be utilized to load: crew option preferences, aircraft configuration, and a priori data (example preset radio frequencies but not code of the day type information).

Additionally, this device would provide backup configuration information to non volatile memories in individual subsystems (example, last configuration prior to a power interrupt) and a failure recording mechanism for GSE.

Only those portions of the flight control subsystem applicable to core avionics are shown in Figure 1. The flight control subsystem has been considered CFE with the exception of a few GFE elements. The GFE assumed is comprised of inertial sensors, flight director backup displays, and the Air Data Computer with backup air sensor instrumentation (Angle of Attack (AOA), velocity, and barometric altitude indicators). This assumption was made based on the differences in basic flight control requirements for various weapons systems.

Subsystem on/off power control and advisory warning displays have been assumed to be CFE. The incorporation of a central clock for distribution to all mission computer digital interface elements is recommended (see 6.1.2.2 and 6.4.2.4). The antenna assets for communications and navigation equipment are routed by GFE equipment recommended in 6.4.2.2. The salient features of the other major components are described in the following paragraphs.

4.3.1 Existing Equipment

A heavy dependence on the use of existing equipment types for the configuration in Figure 1 serves two purposes, namely, allowance for time phasing of individual AVCS equipment and a gradual build up of equipment compatible with the Tactical Information Exchange System (TIES) concept.

Based solely on cost considerations it has been assumed that improvements for functional commonality of Naval avionics equipment will take place in an evolutionary manner. In order to impact any new weapon system the majority of equipment supplied as GFE will have to be "off the shelf" at the time the weapons system is defined. The AVCS program is attempting to stock the

shelves so this can happen in a manner that will result in the use of common units among platforms. Up until the time that all applicable GFE equipment becomes AVCS design compatible, equipment in the inventory now should be utilized as standard equipment. For this reason a key element of the architecture of Figure 1 is the use of Avionics Interface Units (AIU's) which are discussed in detail in 6.4.2.1. The AIU allows modular I/O for each subsystem to accommodate differences in equipment architectural compatibility (i.e., any mix of AVCS or non-AVCS equipment can be accommodated) and the future replacement of existing equipment in a manner that does not directly affect the way the equipment is integrated into the weapon system (i.e., integration is accomplished on the aircraft side of the AIU). The specific equipment items in Figure 1 which are interfaced to the AIU's are arbitrary. The use of the AIU for aircraft interface allows this arbitrary selection of equipment as well as future change to the equipment without affecting the overall architecture or interface requirements. For example, although additional aircraft interfaces will be required for added functions, the direct interface of Joint Tactical Information Distribution System (JTIDS) equipment would be accomplished by replacing the AIU TACAN module. Similarly, replacement of the ARA-63 ILS function by the Multimode Receiver (MMR) would require only an AIU interface module change.

The rationale for using two or more AIU's is to achieve redundancy where it is needed and eliminate superfluous redundancy. This is done by interfacing redundant assets to different AIU's while utilizing only one AIU for non-redundant assets.

Equipment incorporating the standardized interfaces recommended in following sections should minimize the signal conditioning requirements for each equipment AIU module and allow common AIU modules between aircraft in the future. In the ideal case the AIU's could be completely eliminated, however, this is not the likely case because of differences between aircraft. Assuming the AIU's will not eventually be eliminated, they are capable of providing: buffer and distribution of clock signals and aircraft spatial orientation information, bus couplers and terminations, blanking signal logic and distribution; and general equipment interconnects.

Existing equipment is also assumed to be utilized for backup Attitude Reference Indicator (ARI) or Attitude Direction Indicator (ADI) and Horizontal Situation Indicator (HSI) or Bearing Distance Heading Indicator (BDHI) displays.

4.3.2 AIDS Concept Equipment

The architecture of Figure 1 uses the AIDS control and display concept for all core avionics subsystem functions except subsystem power on/off control and Intercommunications subsystem control. The AIDS equipment in Figure 1 is comprised of: the two control and display processors; the primary displays; the avionics controls; and the voice recognition element. The clock is also assumed part of the display and control subsystem. Only core avionics subsystems are considered in Figure 1 however, it has been assumed that the display and control subsystem provides for the requirements of all avionics equipment (including mission equipment). Certain controls for the weapons control and flight control subsystems are excluded because of their peculiar safety requirements.

The prime communications medium between the display and control subsystem and the avionics is a dual redundant time multiplex bus interfaced via the Mission Computer(s). In order to preclude the necessity for separate power supplies for the bus interfaces in the elements to be controlled, individual power on/off controls for each subsystem including the display and control subsystem are suggested. The individual power controls also allow manual control of elements in the case of failures (particularly for bus interface failures). For parallel redundant elements an individual power control for each element that will allow override shutdown of a failed element is recommended.

It has been assumed that due to space limitations, there will be no redundant control panels for cockpit display and control. In order to provide backup for controls, the use of a voice recognition device is proposed. The voice recognition device requires interface with the ICS which makes the ICS

part of the backup path for avionics control, hence, the control of the ICS via the display and control subsystem is inadvisable. For this reason, it is recommended that the ICS have separate redundant controls. This implementation is also consistent with the safety of flight requirements of 4.2, particularly since the ICS requires redundancy and is a necessary element in the operation of UHF and/or VHF communications.

The AIDS concept provides both simplified and redundant/backup control and display for avionics equipment. The concept utilizes multifunction displays and multipurpose control panels. The requirements for wiring through cockpit pressure bulkheads are minimized by utilizing both time and frequency multiplex bus interfaces. All display and control processing is performed within the subsystem by dual redundant processors. Interface with avionics subsystems is accomplished via the same frequency multiplex bus used for cockpit displays and via the mission avionics time multiplex bus. Interface within the cockpit (e.g., throttle and stick controls) are provided via discrete inputs to the multipurpose control panels. Avionics to crew interfaces are provided by combinations of multipurpose television displays, a dot matrix display, a Heads Up Display, a Helmet Mounted Display, programmable legend switches, and a voice recognition element.

The display and control interface in Figure 1 differs from the configuration in the AIDS System Specification (reference h). The system load device interfaces with the Mission Computer(s) in order to facilitate the load of mission variables for all subsystems by one device (in the AIDS System Specification, the load device is interfaced directly to the display processor), however, control of the operation of the system load device is through the display and control subsystem. The interface between the Mission Computer(s) and the display processor utilizes an augmented MIL-STD-1553 time multiplex bus (see 5.3.1.3) along with an external clock input in order to increase the data transmission capability to the displays. The external clock (recommended in 6.4.2.4) is assumed part of the Display and Control Subsystem and would also provide time of day readout. The frequency multiplex bus is a two wire

bidirectional bus (see 5.3.2.2) with frequency assignments established through the Mission Computer(s) in order to allow compatibility with other frequency multiplex bus users. For safety of flight, provisions for discrete warning lights independent of the display and control subsystem are incorporated. The display processor should be capable of alphanumeric indications of voice commands via the cockpit display in the event of a multipurpose control panel failure. It is recommended that a manual ENTER function, independent of the multipurpose control panel, be incorporated with the voice recognition element for backup operation. The voice recognition unit is shown interfaced with the Red side of the ICS hence this unit must meet TEMPEST requirements. Interface with the Black side of the ICS would only be practical if the unit is of a type that can be taught in both plain and cipher mode as part of the preflight procedure.

The AIDS concept equipment forms a key element in the core avionics architecture considered in this report. This display and control concept is fundamental to the architecture because of the ability to provide a consolidated, simple, redundant form of communication between the crew and the avionics. Because of the dependence of other interfaces on the display and control subsystem, this equipment is a prime candidate for GFE. Cockpit display and control has traditionally been under contractor control because of human factors considerations. A reversal of this approach is not recommended, however, the required use of some GFE for cockpit equipment is recommended. The basic AIDS equipment design, along with the incorporation of sufficient display processing capability and the recommended revisions of some interfaces, will allow the flexibility for the airframe contractor to utilize common GFE both in the cockpit and for other crew station controls and displays. Contractor utilization of the GFE controls and displays will require aircraft peculiar software. It is recommended that during AIDS development software applicable to more than one platform be modularized and provided as Government Furnished Software (GFS).

4.3.3 IISA Concept Equipment

The architecture of Figure 1 shows the IISA's interfaced directly to the Flight Control computers. The sensors provide continuous updates of attitude and acceleration information to each Flight Control computer for control of aircraft trajectory. The IISA navigation outputs are interfaced directly to Navigation Processors which provide processed inertial navigation information to the Mission Computer(s).

In the initial generation of the architecture of Figure 1, all IISA sensor interfaces were via the AIU. This interface would have allowed sampling of attitude and rate information for direct use by other avionics. Attitude and/or rate information is required for sensor stabilization, relative navigation calculations, direction finding corrections, antenna polarization control, and data quality checks for imaging systems. This approach was abandoned for several reasons:

- a. The information from the IISA sensors required by other avionics is derivable from the IISA navigation outputs independent of the flight control outputs. The navigation output information requires coordinate transformation for utilization by other subsystems, although this function could be performed within the AIU, its inclusion would be opposed to the intended simplicity of the AIU.
- b. The safety of flight considerations for flight control incorporated within the IISA requires the incorporation of redundancy and strict control of the processing of redundant information in order to maintain capability in the presence of multiple failures. This control function requires a dedicated processor to provide immediate error detection. The inclusion of a dedicated processor within the AIU is not justified particularly when the processing requirements may vary by aircraft type because of aerodynamic performance as well as sensor location.

- c. A direct IISA interface with only the AIU would require an AIU interface with the Flight Control Computer. Thus, the AIU would be an element in series with the flight control subsystem. An analysis of the reliability impact of adding series elements to a redundant system was performed and is presented in Appendix A. This analysis shows the significance of the impact on reliability due to the addition of series elements to a redundant subsystem. The significance of the impact to the flight control subsystem, is increased by the fact that the subsystem is composed of only a few elements and the relationship between aircraft maintenance periods and the high reliability elements that are required for flight control as is shown in Appendix A.

The reliability considerations are the main reason for not interfacing the IISA flight control outputs with the AIU. The same considerations lead to the recommendation that the IISA's be interfaced directly to the Flight Control Computer. The sensor assemblies are good candidates for GFE, however, it is considered necessary that the Flight Control Computer be CFE. The assumed necessity for a CFE Flight Control Computer is based primarily on variations between aircraft aerodynamic surface control requirements, but does not preclude use of GFE hardware for the flight control computer. Although the Flight Control Computer software requirements must vary by aircraft type, it is recommended that during the IISA development program, software applicable to more than one platform be modularized and provided as future GFS.

The redundancy considerations for the processing of IISA flight control information do not apply to the inertial navigation information. The Carrier Aircraft Inertial Navigation System (CAINS II) uses sensor technology similar to that of IISA only for navigation information processing (i.e., CAINS II sensors are not used for flight control). The CAINS II is to replace the AN/ASN-92 GFE inertial navigation system. It has been assumed that future platforms will utilize the IISA attitude and acceleration measurements for both flight control and inertial navigation, hence, CAINS II will not be re-

quired on future platforms. The processing of IISA navigation information has been postulated to be by separate redundant Navigation Processors, however, the processing should be limited to the conditioning of sensor information for subsequent use by the Mission Computer(s) in geonavigation computations. The processed navigation information is provided to the Mission Computer(s) via the dual redundant time multiplex bus for correlation with other navigation information and position and course computations. The redundancy for inertial navigation is a fallout from the use of redundant flight control sensors which are capable of providing navigation quality information, however, the use of redundant Navigation Processors (which could be incorporated in the IISA's without impacting the reliability of the flight control subsystem) can only be justified if they are limited in capability.

The spatial orientation information used for inertial navigation is also required by other subsystems (e.g., antenna stabilization). The orientation information desired by other subsystems are ground track velocity, local altitude, drift angle (difference between heading and ground track), pitch angle relative to an earth stabilized platform, roll angle relative to the aircraft pitch plane, roll rate, and heading rate. In the architecture of Figure 1 it has been assumed that all this information is obtainable from the Mission Computer(s), however, the only mechanism provided to extract this information from the Mission Computer(s) is via a time multiplex interface. Considering parameter rate of change, the sampling of velocity, altitude, drift angle, and heading rate can be accomplished via this interface. For highly dynamic parameters, this interface is inadequate because of information delays. For this reason it is recommended that each Navigation Processor provide for dedicated output of pitch, roll, and roll rate in a standardized serial digital format. In Figure 1 this information is supplied from each Navigation Processor to only one display and control AIU interface module (ie., parallel not dual redundant connection). It is assumed that this information will be combined with sampled velocity, altitude, drift angle, and heading rate information

for general distribution. The sampled information from the Mission Computer should also contain information regarding the age (timeliness) of the data in order to allow its correct interpretation by the various subsystems.

As noted in the previous section, flight control inputs from the crew are not part of the core avionics. Hence, these controls (IISA requires no external control) are not part of the core avionics architecture. The controls for the processing of navigation information in the Mission Computer(s) and pre-flight BIT of the flight control subsystem should be part of the core avionics architecture and processed by the display and control subsystem.

The interfaces for processing flight control and inertial navigation information from the IISA's have been assumed to be dedicated interfaces because of timing constraints and reliability considerations.

In summary, the IISA program has little affect on the core avionics architecture considered in this report. This is true since separate flight control and inertial sensors could provide the same functions, however, the navigation processor (e.g., CAINS II) would need to provide for dedicated spatial orientation information outputs. For reliability reasons the flight control subsystem is considered a stand alone subsystem with the exception of pre-flight BIT controls and information displays. The inertial navigation subsystem is a two level subsystem with sensor and first level processing provided by GFE and correlation and display controlled via the CFE Mission Computer(s). The IISA program is considered a good opportunity to provide standardized GFE for attitude and acceleration sensors as well as some modularized GFS. The use of limited capability GFE inertial navigation processors is also recommended in order to provide mission equipment interfaces and simplify procurement for the navigation quality sensors. Despite the standalone nature of this equipment, utilization of the standard interfaces of 6.2 is recommended for IISA equipment in order to provide for interaircraft compatibility.

4.3.4 Expansion to TIES Concept

The TIES equipment concept is basically a separation of "front" and "back" end processing capabilities for the purpose of allowing functional commonality and flexibility in the use of assets. The concept utilizes common transmitter and receive assets as well as common signal processing elements arranged to provide maximum flexibility in the association of transmitter and receiver assets with signal processing assets. The incorporation of the TIES concept requires replacement of existing equipment for the functions to be performed.

Figure 2 (provided as a foldout at the end of this report) represents an interim step between the use of dedicated existing equipment and its replacement by TIES equipment. All elements below the AIU's of Figure 2 are identical to those of Figure 1. Only the radio communications and Digital Data Link (DDL) assets above the AIU's have been modified in Figure 2. The auxiliary UHF radio has been deleted and the "front" and "back" ends of the radio's and DDL have been separated. (The ARC-159 used for UHF #2 in Figure 1 has been replaced in Figure 2 by assets similar to ARC-182). This separation allows use of common receiver transmitter ("front" end) hardware for both radios as well as for the DDL. The "back" ends for both radio's are also common, however, a separate signal processor for the DDL is assumed. The use of an Intermediate Frequency (IF) switching matrix (see 5.2) allows reconfiguration of transmitter and processor assets to perform DDL, voice, guard monitor and ADF with fewer assets (deleted ARR-69) and more flexibility in terms of fault tolerance. The configuration in Figure 2 is recommended as an interim configuration for any near term aircraft avionics design. This recommendation is based on the additional capability achieved with fewer assets as well as compatibility with incorporation of the TIES concept.

Figure 3 (provided as a foldout at the end of this report) is the same basic architecture as Figure 1 with TIES equipment incorporated. All elements below the AIU's of Figure 3 are identical to those of Figures 1 and 2 with the

exception of the mission computer time multiplex bus interfaces. In addition to the deletion of the auxiliary UHF receiver the AIU interfaces for VHF/UHF #1 and #2, DDL, IFF, TACAN/JTIDS have been deleted in Figure 3. These functions will be provided by TIES equipment and since AVCS standardized interfaces for TIES equipment have been assumed the AIU is not required for direct interfaces associated with these functions. A new AIU module for interface of TIES equipment and the KIT-1A would be required.

The transmitter receiver assets similar to ARC-182 assumed in Figure 2 as well as the antenna interfaces for these units remain identical for the TIES incorporation shown. The "back" ends for the radio and DDL are replaced by redundant two channel TIES narrow band converters and TIES data processors. The IFF and TACAN "front" end assets are replaced by TIES Lx Band assets (a single transmitter and a four channel receiver) with the "back" end function replaced by a non redundant wide band converter and a JTIDS wide band security unit.

The IF switching network of Figure 2 is replaced by frequency multiplex interfaces with the TIES incorporation. The frequency multiplex interface allows greater flexibility than the IF switch configuration (example, simultaneous transmission of one signal by two transmitters) and addition of elements without complicating the configuration. In Figure 3 the frequency multiplex bus provides simultaneous signal interface for all VHF/UHF and Lx band assets. In general, the frequency multiplex bus provides signal interface for all TIES assets (example HF) not just those illustrated in Figure 3.

The TIES configuration in Figure 3 differs from the current laboratory demonstration model of TIES. The frequency multiplex bus in Figure 3 is a two wire bidirectional bus. The TIES laboratory bus is physically similar in that it is two wire, however, signals are coupled on only one wire depending on the desired bus transmission direction. The bidirectional two wire bus is recommended (as discussed in 5.3.2) to make bus routing and equipment location independent and for compatibility with the recommended AIDS bus. The recommended bus is electrically different than the TIES laboratory bus. A redun-

dant frequency multiplex bus configuration is utilized in Figure 3 with only that equipment which requires redundancy (communications related assets) having redundant connections. Combined filter amplifier assemblies in series with the frequency multiplex bus are incorporated for Red/Black compatibility as discussed in 5.3.2.2.

The current TIES concept does not utilize antenna switch networks and would place transmitter receiver assets physically close to the antennas to decrease the transmitter receiver requirements by eliminating most of the coaxial cable losses. The configuration shown in Figure 3 is recommended because it: (a) is compatible with growth into the TIES concept, (b) would not require individual upper/lower antenna front end assets, and (c) does not require individual antennas to be associated with each transmitter/receiver.

The narrow band converter units in Figure 3 assume two channel simultaneous capability. the TIES laboratory equipment utilizes one or three channel equipment. The minimum capability recommended is two simultaneous channels. The Data Processors utilized for TIES equipment should be capable of providing one redundant analog audio output for distribution by the AIU to the ADF antenna control and the ICS. The redundant audio output for the wide band converter data processor is not required, however, identical hardware for the data processors is recommended. A sidetone signal for voice transmission can be provided synthetically by the TIES data processor. This eliminates the necessity for using a second frequency multiplex bus channel to return the audio from the transmitter receiver assets, however, a BIT check of the transmitted audio should be required and allow the generation of synthetic side-tones only when acceptable transmission actually occurs. A nonredundant, serial data interface between the wideband data processor and an AIU display and control interface module is also recommended to provide spatial orientation information for JTIDS relative navigation computations. The interface between the data processor and the display and control module would also provide for deck edge digital data input to the data processor.

The interface of TIES Data Processors to the Mission Computer is assumed to be via the redundant augmented MIL-STD-1553 bus (the wide band data processor does not require redundant connection). The interface of the TIES transmitter receiver assets and the Mission Computer is assumed to be via a redundant connection to only the serial portion of the mission computer bus. The control interface for the transmitter receiver assets and the data processors utilizes IEEE-488 in the laboratory configuration. The 1553 configuration is recommended in order to minimize the number of interface types and provide compatibility. The 1553 interfaces would provide data exchange over the augmented bus and controls including frequency multiplex bus channel assignments over the serial portion of the augmented bus.

The TIES concept offers a significant opportunity to achieve functional commonality while reducing the total number of equipments yet providing redundancy as well as backup capability. The TIES concept also simplifies and reduces the avionics architecture interfaces (illustrated by the reduced number of AIU interface modules in Figure 3). The difficulty of implementing the TIES concept is that it requires a complete revision of standard equipment. The incorporation of the frequency multiplex bus along with AIDS and the establishment of TIES compatible VHF/UHF transmitter receiver assets will greatly relieve the impact of TIES incorporation. It is recommended that the IF access to VHF/UHF assets be incorporated independent of the TIES development, and that a parallel development of GFE Lx band assets compatible with both the JTIDS and the TIES equipment be pursued. Additionally, GFE development of TIES signal processors is recommended to allow incorporation of the overall TIES concept in future platforms.

4.3.5 Compatibility with Mission Equipment

The architecture of Figures 1-3 does not represent any weapons system avionics architecture. This is true because of the differences in requirements for various aircraft. For utility and land based aircraft, additional navigation aids and communications subsystems are usually provided. For non

CV aircraft the ACLS subsystem is not required and inertial navigation subsystems are not incorporated in some aircraft. Most aircraft carry a large complement of additional equipment required for the performance of specific missions. For the architecture of Figures 1-3 to qualify as a Multiplatform Avionics Architecture (that portion of the Weapons System Avionics Architecture that is applicable to many weapons systems), it must be compatible with the variations required by the individual weapons system configurations.

With the exception of the Mission Computer(s), the Display and Control subsystems, and the distribution of spatial orientation information by the Navigation Processors, the deletion of equipment does not affect the Multiplatform Avionics Architecture.

The equipment required in addition to that incorporated in the architecture of Figures 1-3 must be accommodated. This equipment falls into several categories.

- a. Supplemental Core Avionics - Additional elements with functions similar to the core avionics functions are or will be utilized in some aircraft, for example, HF communications, additional Digital Data Links, Global Positioning System (GPS), Doppler Navigation or Correlation Velocity Sensor (CVS), and miscellaneous navigation aids (OMEGA, LORAN, Low Frequency/Automatic Direction Finding (LF/ADF), VHF Omni Range (VOR), VOR Localizer (VOR/LOC), Marker Beacon, and Glideslope).
- b. Weapons Control - Most Navy aircraft incorporate a Weapons Control Subsystem. These subsystems vary by aircraft mission and include control functions for a variety of ordnance. The functions may include elements to provide flight control for weapons delivery or ground controlled bombing, laser trackers for weapon delivery, or sonobuoy launch control.

- c. Generic Avionics - Many Navy aircraft carry generic equipment for the collection and processing of information from a non cooperative external environment. These equipment items include Radar for targeting, weather avoidance, and navigation; IFF interrogators; Electronic Warfare (EW) Equipment for threat warning, threat deception, and target identification; and FLIR (Forward Looking Infrared Receiver) for targeting and threat identification.
- d. Specialized Subsystems - Mission specific equipment is required in specialized aircraft for magnetic anomaly detection, Acoustic data collection and processing, imagery, target designation, Radio Frequency (RF) intelligence gathering, standoff jamming, and command communication links.

No aircraft requires all the additional equipment mentioned above. Nor could any specific configuration of core avionics reasonably accommodate all the equipment because of the level of integration required between the core and mission avionics. Any Multiplatform Avionics Architecture must, however, be compatible with allowing adaptations for different configurations of the mission equipment above. In general, the mission equipment requires integration with the Display and Control subsystem, the Navigation subsystem, the Mission Computer(s), the Flight Control subsystem, the ICS, and the Communications subsystem (i.e., virtually all the core avionics elements).

In the integration of mission equipment and core avionics, several general philosophies are recommended.

- o In order to preclude the incorporation of excess capability (mission equipment peculiar interfaces) in core avionics equipment, the direct interface of core avionics equipment and the mission equipment is not recommended. The core to mission avionics interfaces should

be accommodated through AIU interfaces. An exception to this recommendation would be for multiplex interfaces where information distribution is controlled via the Mission Computer(s) (e.g., time or frequency multiplex bus).

- o For the purpose of achieving greater commonality among mission avionics equipment, the standard interfaces recommended for core avionics should be utilized for mission equipment as well. The use of AIU interface modules or incorporation of separate mission avionics AIU's for mission equipment is encouraged in order to allow the separation of aircraft integration requirements and specific equipment requirements. The direct interface of mission equipment and CFE, as opposed to an interface via an AIU, is however, encouraged for equipment peculiar to a single aircraft.
- o For the purpose of avoiding excess processing capability, mission equipment that operates primarily as a general purpose sensor (e.g., Electronic Support Measures (ESM) receivers or Doppler Navigation) should be interfaced via an AIU with sensor capability on the GFE side and correlation processing (e.g., Radar track and ESM correlation or dead reckoning navigation) performed on the CFE side. This recommendation will allow the CFE equipment to control the distribution of sensor information and provide for signal conversion within the AIU (e.g., synchro to digital conversion for distribution of ground speed and drift angle from the Doppler Navigation equipment) but does not preclude GFE processing elements interfaced to the CFE side of the AIU (e.g., GFE ESM/Radar target correlator).
- o Mission equipment that requires direct CFE interface (e.g., fire control radar) and interface to core avionics (e.g., blanking to and from the radar beacon) should be accommodated via a prime CFE module of the AIU for access to the core avionics interfaces (e.g., display and control module). This recommendation is made to insure that

differences in the signal distribution requirements among aircraft do not require tailoring GFE AIU modules to individual aircraft requirements.

In order to examine the compatibility of the architectures of Figures 1-3 with the accommodation of the mission equipment, some suggested methods of integration have been generated and are given below. A variety of equipment was examined to determine compatibility and not to recommend any particular configuration. A number of navigation aids are included in the examination to accommodate aircraft that may not incorporate inertial navigation. The requirements for these navigation aids, in conjunction with the architecture of Figures 1-3, should be examined carefully because of the redundant inertial navigation capability assumed in Figures 1-3. For example, assuming judicious power distribution, realignment of one inertial subsystem by the redundant subsystem is possible even after a power interrupt, thus the necessity for a large number of additional nontactical navigation aids becomes questionable. In all cases (i.e., with or without inertial navigation capability), navigation equipment should separate the current position sensing function and the navigation processing to allow processing for one navigation solution (course) based on a variety of position information inputs.

4.3.5.1 Supplemental Core Avionics

- a. Communications. The incorporation of HF communications and additional digital data links can be accommodated by utilizing AIU interface modules for the configurations of Figures 1 or 2. The AIU aircraft interfaces allow direct access to the Mission Computer(s) and the ICS for these equipment items. The IF switching matrix of Figure 2 is compatible with these assets, assuming only one additional IF interface is required. If more than one additional IF interface is required, the implementation of the TIES concept (Figure 3), rather than a more complex switching arrangement, is recommended. The implementation of the TIES concept could allow addition of

only front end assets for HF capability while simultaneously providing additional data link capability via the converter and data processor assets.

- b. Civilian Navigation Aids. The incorporation of Very Low Frequency (VLF) and Low Frequency (LF) equipment (e.g., OMEGA, LORAN, LF/ADF) is required in some extended mission aircraft. LORAN is considered a specialized subsystem in that it requires manual operation and, because of potential discontinuance of service, will not be considered further. LF/ADF equipment is likewise considered a specialized subsystem. However, an AIU interface module could be utilized for control, and/or audio and display interfaces via the core avionics. For incorporation of OMEGA navigation, the use of an AIU interface is recommended. However, only position fix outputs from the OMEGA equipment are recommended. Dead Reckoning processing and way point computations should be processed by the Mission Computer(s).

The incorporation of VHF equipment (VOR, VOR/LOC, Marker Beacon) to provide for compatibility with civilian navigation aids and civilian ILS can be accommodated by use of AIU interface modules. These modules would convert display information for routing to the display and control subsystem via the Mission Computer(s) or provide direct interface to backup displays and advisory/warning indicators. Similarly, a UHF Glideslope Receiver can be accommodated. The equipment required for civilian compatibility is not generally utilized in Navy aircraft because of the additional hardware required to perform these functions.

Replacement of the ARA-63 by the Multimode Receiver for the ACLS function also provides the capability for processing civilian ILS signals. The interfaces for the MMR civilian ILS outputs would be provided via the ILS and Display and Control AIU interface modules. Access to VHF/UHF antenna assets for the MMR is available through

antenna switching unit spare inputs (spares available since both units identical) although incorporation of a separate Marker Beacon antenna would be required. Compatibility with the Federal Aviation Administration Microwave Landing System (MLS) can be provided by the MMR but would also require additional antenna assets. With the incorporation of TIES equipment, the majority of assets required for civilian VOR and ILS are also available. In Figure 3, assuming the ARC-182 type transceiver is utilized for separate voice communication, additional assets for processing either VOR or VOR/LOC and UHF Glideslope information are provided by the other two ARC-182 type transceivers. In order to incorporate the Marker Beacon signal an additional VHF asset would be required. Since this asset is provided by the MMR, and the MMR will be compatible with the MLS, the processing of all civilian ILS signals by the MMR rather than the TIES equipment is recommended. However, if a hardware savings in the MMR can be achieved use of the ARC-182 type assets by the MMR should be investigated. The access to the assets could be provided via the frequency multiplex bus of Figure 3. The MMR access to the VHF/UHF assets via the IF switches of Figure 2 is not recommended since the addition of two connections would overly complicate the switching requirement.

- c. Military Navigation Aids. Doppler Navigation and CVS equipment are supplemental core avionics but are considered as mission equipment because this capability is not incorporated in all platforms. In the architecture of Figures 1-3, navigation computations are assumed to be a Mission Computer(s) function. A typical Doppler Navigation subsystem measures ground speed plus drift angle and computes a navigation route. It is recommended that this type system be treated as a GFE sensor interfaced to an AIU with processing provided on the CFE side of the AIU. This configuration eliminates excess redundant processing capability, allows correlation of all navigation sensor information in the processing element, and reduces the number of

different types of navigation display formats. It is recommended that GPS equipment be interfaced in the same manner (i.e., a GFE navigation sensor interfaced to the AIU with navigation course information processed in a central navigation processor on the CFE side of the AIU). JTIDS is not a navigation subsystem, however, it does perform relative navigation calculations. The interface of JTIDS via an AIU is recommended because of the fact that JTIDS communications and TACAN functions are part of the core avionics. The AIU interface will also allow aircraft information (e.g., velocity and altitude for relative navigation processing) to be provided to GFE JTIDS equipment in a standard, real time format by the AIU. The performance of relative navigation computations within the JTIDS equipment is justified since real time computations are necessary in order to control message traffic. The computation for geonavigation within JTIDS is not recommended for the same reasons given for Doppler or CVS, and GPS (central processing for correlation and simplified display). Another justification for not performing navigation processing within JTIDS or GPS is that these equipment items eventually be replaced by TIES (extended Lx-band capability for GPS incorporation) and navigation processing when incorporated in a central navigation processor would not have to be replaced. However, if incorporated within JTIDS and GPS, navigation processing within TIES would have to be redeveloped.

4.3.5.2 Weapons Control

The Weapons Control Subsystem is not shown in Figures 1-3 since it is assumed to have direct interface with only the Mission Computer(s) and Flight Control subsystems which are both assumed to be CFE elements. Display and control, except for controls with peculiar safety requirements, would be provided by the display and control subsystem via the Mission Computer(s). Interface of the Weapons Control Subsystem and other mission avionics would either be direct (e.g., Sparrow Illuminator mode control) or via the Mission

Computer(s) (e.g., ground speed and drift angle from the Doppler Navigation). Interface with core avionics would be via the Mission Computer(s) (e.g., altitude or DDL information) except for discrete outputs from the AIU display and control interface module which is a CFE module (e.g., weapon release signals derived from the DDL in ground control bombing). Since the Weapons Control Subsystem is assumed to interface directly with only CFE equipment or other mission equipment, the core avionics architecture of Figures 1-3 is assumed compatible with this subsystem.

4.3.5.3 Generic Avionics

The control and display for Radar, FLIR, and EW equipment is directly compatible with AIDS concept equipment. Radar and FLIR displays should be provided over the frequency multiplex bus. EW equipment displays are symbolic and therefore can be accommodated under Mission Computer(s) control over the augmented time multiplex bus to the Control and Display Processor. The EW interface with the augmented mission computer bus will facilitate direct memory transfers between equipment elements (e.g., threat file exchange between warning system and jammer) and the loading of a priori data from the Preflight Load and Memory Device. The Radar/IFF Interrogator and/or FLIR require only connection to the serial portion of the Mission Computer bus for control and frequency multiplex bus channel assignments. Because of potential uses for EW video signals by other subsystems and the necessity to continuously update these equipment items due to threat changes, an interface using an AIU module is recommended. Blanking signals would be interfaced through the EW AIU modules and through the display and control interface module for the Radar and IFF interrogator.

The instantaneous field of view used by Radar and FLIR sensors requires information about the aircraft spatial orientation. In the architecture of Figures 1-3 this information is assumed available for highly dynamic parameters (roll, pitch, roll rate) from the Navigation Processors, for sampled

parameters (velocity, altitude, drift angle, heading rate) in the Mission Computer, and in a combined format from the display and control AIU interface modules. The recommended interface is through the AIU.

The recommended Radar/FLIR interfaces are directly compatible with image correlation processing in that standardized format display information is available over the frequency multiplex bus independent of the display currently presented. Additionally, the Radar/FLIR sensors would operate utilizing the same aircraft spatial orientation information. Electronic Warfare sensor information as well as IFF interrogator response information would also be available over the time multiplex bus for additional correlation and target identification. Subsequent display of correlated information would be accomplished over the frequency multiplex bus.

The interfaces shown in Figures 1-3 for existing core avionics (items above the AUI's) and the AIU's are dedicated interfaces. Multiplex interfaces are not required and are not provided for in the equipment shown (ARC-182 does provide a 1553 interface). Some of the mission equipment recommended for AUI interfaces do require multiplex interfaces. The recommended configuration for some equipment (e.g., EW equipment) interfaces mission equipment directly to the augmented mission computer time multiplex bus. Other equipment (e.g., Radar) are recommended to interface to the serial portion of the mission computer time multiplex bus and to the frequency multiplex bus. Direct interface to the mission computer time multiplex bus is recommended, however, there are problems with this approach. In Figure 3 there are twenty terminals connected to the mission computer bus which has a maximum capacity of 31 addresses, thus there is a limit to the number of additional terminals that can be utilized. Since existing equipment will also be utilized for mission avionics the problem of MIL-STD-1553 A vs B arises. The resolution of how many terminals and the configuration of different types (A and B) are assumed part of the CFE Mission Computer(s) arrangement, however, the Mission Computer interfaces in the AIU's which are also CFE can be utilized to provide a buffer between A and B terminals or an extension of the mission computer bus.

4.3.5.4 Specialized Subsystems

As discussed in section 3.0, the peculiar requirements of specialized subsystems should not be accommodated in a Multiplatform Avionics Architecture. The incorporation of redundant interfaces between virtually all equipment on the CFE side of the AIU in Figures 1-3 and the use of at least two AIU's, should provide adequate survivability considerations for specialized equipment. The control of time and frequency multiplex interfaces by CFE equipment as well as CFE software in the display and Control Processor should allow sufficient flexibility to incorporate some of the Specialized Subsystems functions (e.g., display and control) within the architecture. The use of AIU's for distribution of blanking and aircraft spatial orientation signals through CFE display and control modules should likewise accommodate the specialized subsystems.

5.0 INTERFACES AND INTERFACE REQUIREMENTS

The avionics architectural design determines the interfaces between each element of a subsystem and all the other elements of the weapon system. Reducing the impact of implementing the interfaces to both the weapon system and the subsystems in terms of size, weight, power, cooling, and complexity while maintaining commonality, growth capability, and technology transparency is a natural method to achieving the goals of the AVCS program. To this end, interface requirements were examined in some detail.

For avionics equipment, the required interfaces fall into three general categories.

- a. Control/Status Information - operating commands or controls, timing or feedback information, status or control outputs.
- b. Information Transfer - of either raw, partially processed, or directly interpretable data.

c. Resources - power and cooling.

The requirements for resource distribution were not considered as part of this study. The types of interfaces required for control/status and information transfer can be grouped as follows.

- a. Binary - discrete signals (on/off, BIT command, light outputs, status I/O, timing signals, etc.) where state as a function of time is the information conveyed. Although grammatically incorrect and not recommended, tri-level signals (three logic states) would be considered in this category.
- b. Digital - signals where the sequence of states over a fixed period of time convey coded information (e.g., digital bytes or words).
- c. Analog - time varying signals below 20 KHz or signals where a continuously variable status is conveyed instantaneously (e.g., audio signals, potentiometer controls, thermocouple outputs, feedback signals, synchro signals, etc.)
- d. Video - signals above 20 KHz where amplitude and time are used to convey information (e.g., display signals, detected radar signals). For the purposes of this study, audio signals are not considered in this category and are classified as analog signals.
- e. Modulated Carriers - signals where the information in any form is mixed with a known continuous signal for the purposes of transmission (e.g., IF signals or RF signals transmitted or received via an antenna).

Binary and analog signals are used primarily for status and control information. Video and modulated carrier signals are used primarily for the

transfer of information. Digital signals, because of the coded characteristics, may be used for both purposes. The requirements for each signal type are dictated by the individual subsystem. The manner in which the interfaces can be implemented is primarily a function of timing relationships.

Signals may be required continuously, aperiodically, or with some regularity over time. Additionally, the signals may for reasons other than information content be required to be available on a continuous basis (e.g., safety of flight considerations). Signals that are required continuously, or to be available continuously, and signals with strict time dependencies need to have dedicated interfaces. Dedicated or continuous interfaces for all signal types are required, however, multipurpose interfaces, where applicable, offer significant advantages to weapon system avionics. The obvious advantages are weight, size, and power reductions. Somewhat less obvious are increases in: reliability and maintainability (by reduction in wire counts); the degree of achievable system integration (by more access to information); and the growth capability (due to the flexibility of multipurpose interfaces). Multipurpose interfaces fall into two categories switched and multiplexed.

Switched interfaces for signals needed aperiodically provide a reconfigurable arrangement of dedicated interfaces and except for the switching requirements, will be treated as dedicated interfaces. Multipurpose dedicated interfaces (e.g., a control line, which has a different meaning during BIT) which may be considered a form of switching accomplished internal to the subsystem will be considered as dedicated interfaces.

The ability to multiplex signals is dependent on the multiplex technique and the signal type. Any of the signal types converted into a modulated carrier (by AM, FM, PM, FSK) are compatible with frequency multiplexing. The time dependent nature of binary, analog, video, and modulated carriers is in general incompatible with time multiplexing. The property of digital signals that information is transferred over a fixed period of time, make these signals compatible with time multiplexing.

Point-to-point signals that are multiplexed by a subsystem to increase the information transfer on a single interface (e.g., interleaved roll and pitch attitude information) will be considered a single signal. Multiplexing in the context of this report is a technique used for distribution purposes.

Each of the interface types is discussed individually in the following sections along with the distribution media for the signals, i.e., dedicated, switched, multiplexed. In the discussion, negative comments regarding current practices or deficiencies in current techniques are italicised. For each italicised comment a positive recommendation is given in Section 6.0.

5.1 DEDICATED INTERFACES

Dedicated interfaces are required in a number of different applications, e.g., for signals with time critical requirements, for control over probability of false alarm, for safety of flight, and for signals with a large information content (broad bandwidth or a high data rate). Dedicated interfaces for binary, digital, analog, video, and modulated carrier signals are necessary to accommodate general weapon system avionics requirements. *There are no military standards for general purpose dedicated interfaces.* DOD-STD-1399B recognized the need for such standards for shipboard systems, however, to date only requirements for the use of standard components and the documenting of electrical interfaces have been published.

Standardization of dedicated interfaces is necessary in order to simplify integration efforts, to meet new requirements, and to preclude the necessity for adaptation of GFE to the weapon system avionics architecture.

5.1.1 Binary Interfaces

Binary interface requirements are similar to those of digital interfaces except in the criticality of timing and/or in their electrical format. Dedi-

cated binary interfaces that are not constrained by electrical format requirements (i.e., discrete I/O signals) will be treated as digital signals (for example, status/mode/command signals and timing signals) since the physical interface requirements are the same. Only two types of electrically constrained binary signals should be required in future avionics subsystems. Subsystem on/off power inputs will be required and, in cases where safety of flight is concerned, dedicated status and warning outputs are necessary.

5.1.1.1 Power Input

Subsystem power, even though controlled by a power distribution subsystem, will require a subsystem on/off interface. Historically this interface has been accomplished by several methods: switch control of the input power leg, switch control of the return leg (switched ground primarily used for 28 V dc input), and switch control of a signal (primarily switched 28 V dc power or ground leg) used to operate a subsystem relay for application of 115 V ac. With the advent of sophisticated power distribution subsystems (Advanced Avionics Electrical System (AAES) for example) only the equivalent of switch control of the prime input power leg should be required. Although the incorporation of AAES will provide switching of input power in the aircraft, the location of switching used in the interim will not have impact on AAES incorporation. Switching located in the aircraft would be replaced by the new power subsystem. The conversion from 115 V ac 400 Hz to 270 V dc prime power would require replacement of power supplies in individual equipment and thus any switching incorporated in the power supply would be removed via the replacement.

In order to allow simple and safe cockpit/station controls, it is recommended that all power interfaces be required to operate via a switched ground return to 28 V dc (maximum current 0.2 ampere) which controls the input power by switching within the subsystem. This may necessitate additional hardware (relays) in each subsystem, however, it will provide uniform cockpit controls which will aid in the conversion to an AAES type subsystem. The additional

relays can be removed when conversion to an AAES type subsystem is made. The necessity for additional hardware will only occur in moderate current 28 V dc subsystems. For safety purposes, subsystems requiring externally switched 115 V ac or high current 28 V dc are normally controlled through relays that are part of the aircraft. The shift in responsibility for power relays to the subsystems is not overburdening, particularly when safety, reliability, and commonality are considered. Additionally, this shift would reduce the impact of AAES on existing platforms.

5.1.1.2 Warnings

Control and display of cockpit/station warning lights/tones, etc., are readily amenable to digital multiplexed interfaces. However, since the functions for which these indications are required sometimes involve safety or safety of flight considerations, it is not recommended that all such warnings be processed by multiplexing.

It is recommended that all safety or safety of flight warning outputs be of a current sink type (maximum of 0.2 ampere from 28 volts). This interface is compatible with direct control of instrument panel warnings and is also compatible with light dimming circuitry. Flashing or blinking of warnings should not be accomplished through this interface. Warnings requiring flashing or blinking should be output via a continuous dedicated binary interface with the on/off warning timing controlled in the instrument panel. This approach prevents annoying asynchronous warnings and allows synchronization with audio warnings.

5.1.2 Digital Interfaces

Any type of electronic, as opposed to electrical, binary signal requiring point-to-point interface will be considered a dedicated digital interface. Digital information is in general a coded format and the compatibility of format as well as physical signal type is necessary for true interoperability.

For dedicated digital signal interfaces, standardization of format is considered impractical, however, it is practical to standardize the physical parameters of the interface. For discrete I/O signals only standardization of physical parameters is required.

The data rate requirements for digital interfaces as considered in this report vary from 10^{-4} to 10^{+6} baud. Discrete I/O signals may not change state for periods of hours, (e.g., subsystem mode control). Serial digital data transfers may operate at megabit rates continuously. Although the requirements for digital signals may vary significantly, since the majority of dedicated digital interfaces will be necessary because of the criticality of timing or high data rate considerations, a single standard is considered prudent.

5.1.2.1 Existing Standards

There are a number of standards and specifications regarding digital interfaces, however, with few exceptions a single standard/specification does not comply with the range of requirements of the others.

MIL-STD-188C is applicable to military communications systems. For digital signals standards for low level (± 6 V one wire) and high level (± 60 V one wire, not recommended) were established. The applicable requirements of MIL-STD-188C are to be replaced by MIL-STD-188-100 for common long haul and tactical communications systems and MIL-STD-188-200 for tactical communications systems. MIL-STD-188-200 has not been issued. MIL-STD-188-100 describes balanced voltage (± 3 V on two wires) and unbalanced voltage (± 6 V one wire) low level interfaces and repeats the not recommended high level interface of MIL-STD-188C. MIL-STD-188-114 supersedes the low level interface requirements of both MIL-STD-188C and MIL-STD-188-100.

FED-STD-1020 and FED-STD-1030 adopt EIA-RS-422 (balanced voltage) and EIA-RS-423 (unbalanced voltage) interfaces respectively in their entirety for telecommunications equipment. MIL-STD-188-114 is basically a restatement of

EIA-RS-422 and EIA-RS-423 and hence is compatible with FED-STD-1020 and FED-STD-1030 with one important exception. In order to allow for common mode voltage (i.e., induced potential on both leads) for the balanced voltage interface, the requirements for EIA-RS-422 line driver outputs are not stated relative to a ground potential. MIL-STD-188-114 requires (indirectly) that the signals on the two lines of the balanced interface be of opposite polarity relative to ground in order to maintain compatibility with the unbalanced MIL-STD-188-100 and MIL-STD-188C receiver interfaces (i.e., each balanced output line must swing above and below ground for compatibility with the older +6V one wire receiver). As described in MIL-STD-188-114, the only electrical incompatibility between any combination of balanced or unbalanced receivers or line drivers for any of the low level MIL-STD-188 or federal standards is that the FED-STD-1020 balanced generators cannot drive a MIL-STD-188C or a MIL-STD-188-100 unbalanced low level receiver. There is also a convention incompatibility with MIL-STD-188-100 in that the meaning of 1 and 0 is reversed between MIL-STD-188-100 and MIL-STD-188-114. *The rationale for requiring electrical compatibility for a mixture of balanced line drivers and older unbalanced line receivers, when sign compatibility is not achieved in any combination between MIL-STD-188-100 and MIL-STD-188-114, does not make sense, particularly since the implementation of MIL-STD-188-114 balanced interface requires dual voltage power supplies and the implementation of FED-STD-1020 (or EIA-RS-422) does not.*

For other than telecommunications equipment, there is no general military standard. Each equipment utilizes its own interface which is not necessarily compatible with other equipment, (e.g., AN/AYK-14 Discrete Interface Module utilizes EIA-RS-422 and EIA-RS-423 compatible interfaces, whereas the PROTEUS module utilizes MIL-M-38510/104-03/04 line drivers and receivers which should be compatible with EIA-RS-422 but are not required to be, while each of the four Navy Tactical Data System (NTDS) interface modules is different from the other and none is compatible with EIA-RS-422). MIL-STD-1397 describes the four NTDS interfaces for use on ships, however, since none of the four is the

same as those of the MIL-STD-188 series or other general avionics interfaces they are not considered a basis for an avionics equipment standard.

None of the existing standards adequately address requirements for other than single point-to-single point dedicated digital interfaces. In the practical application of digital and binary interfaces fail safe provisions are required (i.e., proper input state sensed when interconnecting equipment is missing or turned off). In order to justify requiring standard interfaces they must be compatible with future integration as well as equipment interchangeability. Even when standard interfaces are employed, design practices used for fail safe operation and line termination may limit the ability to further integrate subsystems. This is so because the line drivers are limited by the fail safe and line impedance terminations in the number of receivers they can drive.

The EIA standards have been modified. The latest versions are RS-422A and RS-423A which incorporate minor requirements for additional, although not complete, compatibility with International Telephone and Telegraph Consultative Committee Recommendations. *In the revised EIA standards, the problem of multiple receivers and line terminations is recognized but not adequately addressed.*

5.1.2.2 Future Requirements

The dedicated, point to point transmission of digital information at rates above 10 MHz is not anticipated in the future. The limitation is based on the constraints of signal skewing due to transmission delays encountered in aircraft installations. Transmissions at higher rates are practical only if the clock (sync) information is sent over the same interface. If such interfaces are required, the use of point to point optical transmission would be the most practical medium based solely on Electromagnetic Interference (EMI) considerations. The use of an optical transmission medium for point to point transfer of digital information at or below 10 MHz is in general not justified

since the same function can be performed by a cheaper, lighter weight electrical interface. In specific cases Electromagnetic Pulse considerations may justify such interfaces. Transmission at higher data rates over optical transmission lines for the purpose of replacing multiple electronic interfaces is justified and is discussed in bus requirements.

The EIA-RS-422 and EIA-RS-423 standards can satisfy the anticipated future needs for dedicated digital interfaces assuming that no more than 6 to 10 receivers will ever need to receive the same information from a single line driver (see 6.2.1.2). With few exceptions, this is a reasonable assumption. The only difference between EIA-RS-422 and EIA-RS-423 is in the line driver requirements since the actual line receiver requirements are identical. The EIA-RS-423 line driver is a single wire interface whereas the EIA-RS-422 line driver supplies a differential output on two wires. A number of the applications discussed for digital interfaces do not require balanced voltage circuits (EIA-RS-422), hence use of the single wire unbalanced circuit (EIA-RS-423) would allow a reduction in aircraft wire count and associated weight, along with increased reliability relative to the balanced voltage circuit utilization. Despite the advantages of having two types of interfaces for differing applications, only the use of balanced voltage circuits is recommended.

A general military standard adopting EIA-RS-422 type interfaces for use in all future digital interfaces is recommended. Exclusion of the single wire interface is recommended for two reasons. The interface connector pin limitations on modern avionic equipment will force increased use of other than dedicated digital interfaces where they are not necessary if the balanced interface is required. Secondly, the number of applications where a dedicated digital interface is required and the unbalanced interface is adequate are not significant enough to warrant a separate interface standard. This is particularly true when EMI and the common mode rejection capabilities of the balanced interface are considered.

In order to allow for future integration (as well as the bus interfaces discussed in 5.3.1.3), dedicated digital interfaces should be configured so they are multipurpose. For multipurpose application, tri-state operation of line drivers is required. The use of the tri-state line drivers is only effective when the subsystem containing the line driver is externally controlled (i.e., line driver output is generated in response to an external input which causes the tri-state output to transition from the high impedance to digital output mode). For dedicated interfaces, this type of operation is unusual since it is simpler to let the dedicated line driver free run and have the external control signal manipulate the sampling of the information at the line receiver. Discontinuing this practice and requiring the use of tri-state line drivers wherever possible has the potential of allowing dedicated interfaces to act as busses in future integration efforts.

5.1.3 Analog Interfaces

Analog or proportional interfaces are being replaced by digital interfaces at a rapid rate; however, analog interfaces are not likely to become completely obsolete in the near future. The conversion of a signal to digital format is limited by sampling rate requirements. There are also accuracy limitations to a digital conversion, but for signals propagated through aircraft wiring the quantization error of a digital conversion may be less than the induced EMI errors. Devices that are compatible with digitization will require analog interfaces until the signal reaches the point where the conversion takes place (e.g., voice, potentiometer settings, synchros, temperature sensors, etc.). Complete elimination of analog interfaces would require additional hardware (power supplies, Analog to Digital (A/D) converters) located with remote devices.

There is no strong technical reason for standardizing analog interfaces since they are dedicated special purpose interfaces. The integration of future systems could benefit from standardized interfaces particularly where future technology has the potential of revising the sampling rate limitations of

digital conversions. For this reason, along with the fact that reasonable standards should not have a measurable impact on future subsystems, standardization for amplitude analog and phase analog signals is recommended.

Two standards for amplitude analog signals are recommended to accommodate unipolar (0-10 V dc recommended) and bipolar (-5 to +5 V dc recommended) signals. The recommended signal ranges are arbitrary. However, equivalent peak to peak swings for bipolar and unipolar signals are strongly recommended in order to allow common voltage quantization values for the two types of interfaces.

Two standards for phase analog signals are recommended to allow synchro to digital conversions of back-up flight control displays in future avionics integration. The recommended ranges are those currently utilized, namely 26 V and 115 V levels.

5.1.4 Video Interfaces

Video signals are utilized primarily for detected radar or sync and video display information where signal amplitude and timing convey the information. Approximately 16.9 MHz bandwidth is required for display information. Radar return information requirements are a function of radar parameters. Standardization of both these interfaces can have significant impact in future avionics.

Subsystems requiring cockpit/station display are often procured separately from the display, thus standardization would offer a means of mixing and matching displays and subsystems as well as allowing increased utilization of station displays for different missions. Radar video signals are typically very special purpose in nature; however, electronic warfare and radar altimeter video information have potential application for use by other subsystems. Standardization for radar video signals (17 MHz bandwidth for a 120 nsec pulse) would facilitate future utilization of this information.

Although there is no reason to use the same standards for radar video and display sync and video, the requirements are so similar that there is no reason not to use the same standard and thus reduce the number of different standards. A standard 17 MHz interface for both type signals is recommended. However, care should be taken to have this standard applicable only to signals where both time and amplitude convey information. Signals that are often treated as video (e.g., blanking signals or transmitter pulse modulation signals) do not require amplitude information and should be treated as binary signals.

5.1.5 Modulated Carriers

The requirements for modulated carrier signals include carrier frequencies in the range of 10^3 to 10^{10} Hz and higher if optical transmissions are considered in this category. Interfaces for these signals are usually very subsystem sensitive (e.g., RF transmissions by the subsystem or specialized IF signals). General standardization for modulated carriers is not possible without subsystem performance impact and is not considered prudent.

It is possible to generate a skeletal standard that describes only broad requirements for optical interfaces, however, at this point in the technology, such a standard is considered premature. A preliminary standard describing general requirements of optical fiber connectors has been drafted (MIL-C-85044) and a general requirements standard for fiber optic cables is in effect (DOD-C-85045). General standards for optical interfaces should be delayed until broader application of the technology is achieved.

The first IF frequency utilized in most radio communications equipment is typically 70 MHz. Because of the proliferation of radio receivers and transmitters on a single weapon system (e.g., 2 UHF transceivers each with a separate guard channel, auxiliary UHF receiver with a guard channel, and a UHF digital data transceiver) standardization of the first IF frequency could have significant commonality and reliability impact. If access to the first IF

were available (recommended action to achieve capability in 6.1.2.3.2), the transmitter and receiver assets could be reconfigured by IF switching (recommended development in 6.4.2.3) to perform the required functions, thus achieving greater functional commonality as well as increased reliability. For this reason, a standardized IF interface for radio communications equipment is recommended.

5.2 SWITCHED INTERFACES

As noted earlier switched interfaces are used primarily to provide a reconfigurable arrangement of point to point dedicated interfaces. The switching of interfaces is required for: control of standby redundant assets; configuration or mode change (e.g., ADF vs. omnidirectional processing or secure/clear voice); management of scarce resources (e.g., antenna switching).

The advisability of utilizing switching to control redundant standby assets is highly dependent on the number of failure paths and the reliability of the elements, the switching device, and failure monitors. Since the method of switching implemented affects the reliability of the subsystem and the reason for implementing switching is to improve reliability, it is not considered prudent to place general requirements on redundancy switching.

Switching to accomplish configuration or mode changes may involve any of the signal types discussed above. Multiplexing used in lieu of switching is attractive particularly since frequency multiplexing is compatible with all signal types. The implementation of frequency multiplexing requires a significant hardware investment which is not justified for simple switching requirements. In those cases where multiplexing is not cost effective, continuation of current practices utilizing relay control matched to a weapon system requirements is considered adequate. Relay control should be standardized using the binary interface type discussed in 5.1.1. Since it is assumed that the signals to be switched are standardized, the lack of standardization in switching implementation should have no impact on future integration. This is

true since switching involves only the destination of the signal and integration efforts can utilize the signal source without being affected by the switching.

Another reason for allowing weapon system independent switching is for compatibility with Red/Black requirements. Although not completely resolved, a strict interpretation of Red/Black requirements would require every element on a multiplexed bus (time or frequency) that contained classified information to meet Red requirements. By allowing for independent switching only the elements actually requiring classified information or the switches and controls would need to meet Red requirements.

An alternative to both discrete switching and multiplexing is multiplex switching. This alternative is attractive where frequency multiplexing is not cost effective or where discrete switching is overly complex, e.g., reconfiguration of three or more elements. One application for multiplex switching is for the IF's of radios.

A higher degree of functional commonality for radios could be achieved by control of the IF signals. Guard channels are provided in auxiliary radios (receive only) as well as general radios. Separate UHF transmitters are utilized for digital data and voice often when a redundant voice radio is incorporated in the weapon system. Having the capability to reconfigure radio assets via switching of the IF would allow greater utilization of assets. It would also simplify switching between radio types (e.g., UHF to VHF) and in the processing of ADF functions. For this reason development of a stand-alone, fail-safe, solid state, four by four, 70 MHz multiplex switching network is recommended. The any one of four input to any one of four output arrangement is recommended since it is the simplest arrangement that will allow multiplatform application. The 70 MHz IF is recommended since it is compatible with many existing first IF's. *In order for this development to be maximally effective, the signal requirements for all radios must be standardized or the radios must be made multipurpose.* The multiplexed IF switching achievable with such a unit

would allow simultaneous and reconfigurable operation of resources to perform such functions as voice communication, ADF, guard monitoring and data link.

Switching is required for the utilization of scarce resources. The major application of this type switching is for RF signals to antennas. In this area there is no necessity for all switching to be weapon system dependent, at least for UHF, TACAN, IFF and ADF antennas, since the requirements are common across weapon systems. For this reason development of a standard antenna switching unit is recommended.

5.3 MULTIPLEXED INTERFACES

Both time and frequency multiplexing of signals used in avionics equipment are practical. However, because of timing constraints on other types of signals, only digitally formatted information will be considered for time multiplexing.

5.3.1 Time Multiplexing

The requirement for distribution of specific digital information in a given format is typically only from point to point. Despite this fact, a common method of distribution is by bus, i.e., a common lead(s) with multiple connections that can serve as a point to point interconnection for many elements and, if required, a point to several points interconnection. There have been two major reasons for the application of busses to provide what is essentially point to point communications, namely, as a means of reducing interfaces and hardware and the fact that available point to point links are capable of transferring information faster than the recipient can use it. The reduction in hardware and interfaces is achieved by using one interface in each unit to provide all similar point to point interfaces for that unit. The one interface rationale, when applied to several units with different point to point communications requirements among themselves, will logically lead to a

bus configuration. The bus is feasible only because it can carry more information than the sum of all the users need to receive.

Although not a major technical reason for utilizing a bus, the flexibility in terms of the addition or change of elements in an avionics architecture that is achieved almost without cost by a bus configuration has led to general acceptance of this medium. The wide acceptance of this medium should not overwhelm its technical application, particularly when future technology trends are considered. With the ever-increasing speed of digital processors and the expected impulse to this increasing trend that will be provided by developments such as very high speed integrated circuits (VHSIC), the technical feasibility of continued bus techniques should be examined relative to the flexibility advantages.

A bus is technically justified for a given application only if it reduces hardware requirements and has enough excess capacity to insure that multiplexing does not affect the ability of the recipient to perform the required functions. As the application of digital processing becomes more widespread and the ability of processing elements to assimilate and exchange more information increases, the requirements for bus data rates will also increase. The combined effect of more units interfacing to the bus and the desirability of a larger information exchange will cause the bus data rates requirements to increase dramatically. Only if the emerging technology in the area of busses can achieve the required data rates, and actually utilize fewer resources (hardware and software) than point to point counterparts, can the flexibility advantage of bus techniques be considered cost free.

5.3.1.1 Digital Information Types

Information in digital format that is used by the weapon system avionics has many different characteristics, it may vary from individual bits representing the state of discrete switch positions to complex groupings of bits representing encoded information. For the purpose of this discussion, all

digital information is assumed to be in groups of bits (words) which may contain: multiple pieces of information (binary representation of many individual switch positions), a single piece of information (binary representation of a value or instruction), or less than a single piece of information (encoded video binary string or partial instructions). The types of digital information that need to be considered for avionics subsystems are:

- a. Control Word - digital words that require one or more actions to be taken by the recipient.
- b. Data Word - a digital word that contains information relevant to the operation of the recipient.
- c. String - a group of digital words which contain information relevant to the operation of the recipient but for which individual words contain no useful information.
- d. Block - a group of digital words where each word contains useful information but the entire group of words is required in order to be relevant to the operation of the recipient.

The terms above have precise meanings in many other contexts that differ from the definitions above. The intent of the above definitions is to define the meanings of these terms in the context of this discussion and not to provide universally accepted definitions.

5.3.1.2 Existing Standard

MIL-STD-1553 established requirements for digital, command/response, time division multiplexing (data bus) techniques on aircraft. The requirements are for an alternating differential type transmission medium operating at one megabit rate. Manchester coding of the bits is utilized, hence the bus actually

operates at a 2 MHz rate, i.e., a 50% overhead to allow error detection of bit transmissions. All transmissions on the bus are 20 bit time words of which only 16 bit times contain information. The additional bit times are used for timing synchronization and word parity. Because of the limited number of words in a transmission, no check sum type error detection is required by the standard but its use is not precluded in data transmissions.

The bus utilizes a single (although transferable) bus control scheme where all other elements attached to the bus are considered remote terminals. All communications are controlled by the designated bus control element through the transmission of Command Words and the reception of Status Words from the element receiving the command, (Status Words are not used in broadcast mode - i.e., transmission of commands to all bus elements). The 32 bits of information contained in the Command Word plus Status Word are allocated approximately as follows: 4 bits reserved for future growth or test, 3 bits for fault detection and management, 20 bits for bus management functions, and 5 bits (Subaddress/Mode field) left to the discretion of the individual system designer. Of the 5 bits to be utilized by the system designer, 2 of the 32 combinations are reserved to control the meaning of 5 of the bus management bits (i.e., gain additional control). When the additional control (Mode) is utilized, the additional 5 bits of control (32 states in the Data Word Count/Mode Code field) are used as follows: 17 states reserved for future growth, 9 states for fault detection and management, and 6 states for bus management functions.

The requirements for the 1553 data bus were very carefully designed around error detection as is indicated by the approximately 27% of all bit times in a Command/Status response allocated to fault detection and fault management functions. By the use of two of the 32 states reserved for individual system designers remote terminal controls for redundant busses are provided thus increasing the provisions for fault management.

Of the 20 out of 32 bits (Command Word plus Status Word) used for bus management functions, 10 are used for terminal addressing (5 in the Command Word and 5 in the Status Word). The 5 bits of terminal address in the Status Word are redundant in that their only purpose is to insure that the Status Word originated from the remote terminal to which the Command Word was sent. (In normal operation, this will always be the case.) The 5 bits of address in the Command Word allow control of 30 remote terminals by the bus controller since one address is assigned to the bus controller and one is reserved for the broadcast mode. The provision for 30 remote terminals is sufficient to accommodate almost any realistic arrangement for an avionics system. However, the limitation of 31 total addresses severely limits the ability to utilize the same elements between aircraft or systems since the assignment of addresses must be consistent between aircraft or systems unless some method of reassignment is provided.

The apparent lack of flexibility provided for system design (30 states) is overcome by the use of Data Words which may, in turn, be used as Control Words for individual elements on the bus. The transfer of data words in a string or block is limited to a maximum of 32 words per transfer.

The 1553 standard places no requirements on how the bus will be utilized other than the requirement for compliance with the specified parameters (broad description of electrical interface and low level protocol). The four avionics examples given in Section 6 of reference c (F-16, LAMPS MK-III, YAH-64, and B-52) all utilize a fixed remote terminal scheduling technique where multiples of the basic rate may also be used for timing. This utilization makes sense when consideration is given the fact that the bus control elements will typically be part of a processor that has its own function to perform and only a limited amount of overhead processing can be accommodated. The 1553 was designed to allow maximum processing of low level bus protocol functions in the bus interface without overburdening the processor to which it is interfaced. Higher level protocol processing (e.g., for processor to processor interfaces) is not addressed.

An idealized scheduled system would operate as follows. Assume a base sampling rate of 6.25 Hz and three multiples thereof (example: 160, 80, 40 and 20 msec cycle times) and provision for 30 remote terminals. On a long term basis, depending on the base rate, each terminal would be allowed a maximum of 5333, 2666, 1333, or 666 μ sec of bus utilization per cycle. Cycle times may be different for each remote terminal thus allowing increased bus utilization for some terminals, however, for simplicity, it will be assumed that all terminals are allocated identical bus utilization rates. The simplest information exchange involves one Command Word and one Status Word which requires between 44 and 60 μ sec to accomplish depending on response times. In this period (52 μ sec average), only 5 bits of information are actually transferred thus the information rate is 10.4 μ sec per bit average or 96 kilobits per second (10% of the actual bit rate). Although the overhead appears significant, the actual rates that are achieved are considered more than adequate for status and control functions. When data words are utilized to effect information transfer between remote terminals (RT) and the bus controller (BC), the overhead decreases significantly. For example, a 32 word transfer between the bus controller and a remote terminal requires between 684 and 700 μ sec (average 692 μ sec) with 512 bits of information transferred resulting in a 740 kilobit/sec rate (or 74% of the actual bit rate). It should be noted, however, that this transfer cannot be accomplished in the 666 μ sec (50 Hz rate) cycle. In order to accomplish this transfer in the minimum cycle time, a base rate less than 6 Hz (with a fourth harmonic of less than 47.6 Hz) would be required.

When asynchronous transfers between remote terminals are necessary, a significant amount of overhead is required to notify the bus controller of the transfer, setup and effect the transfer, and check for completion of the transfer. In order to accomplish this transfer, six transactions involving both overhead (7 Command Words, 7 Status Words, and 4 Data Words) and the actual data transfer have been assumed. This overhead requires between 386 and 490 μ sec depending on response times. The transfer of 32 words requires a total of 1026 to 1130 μ sec (average 1078 μ sec).

As noted earlier, the maximum word transfer per transaction is 32 words. If more than 32 words require transfer, several transactions are required. Accommodating asynchronous transfer between remote terminals within a fixed schedule mode of operation would in a single cycle allow: no 32 word transaction at the 50 Hz (666 μ sec/terminal) sampling rate; one at the 25 Hz (1333 μ sec/terminal) sampling rate; two at the 12-1/2 Hz (2666 μ sec/terminal) sampling rate; and four at the 6-1/4 Hz (5333 μ sec/terminal) sampling rate. Except for the case of the 50 Hz polling rate, the remote terminal to remote terminal transfer data rate would be 800 16 bit words/sec maximum. This transfer rate is considered acceptable except when the data transferred is a String or a Block of information. As defined earlier, the recipient of String or Block information cannot complete utilization of the information until the entire transfer is complete. Thus moderate block sizes (1024 words) would require more than 1 sec to transfer and would necessitate a corresponding delay in processing.

There are a number of methods to get around this problem, for example, reducing the number of terminals to be serviced thus increasing the sampling cycle time for each serviced terminal, using variable cycle times for different terminals, or using an auxiliary 1553 bus dedicated and interfaced to only the terminals requiring asynchronous transfers. All of these approaches can perform adequately. However, if they are implemented the flexibility in terms of growth potential is not achieved since the bus implementation is tuned to its initial design. In the case of the auxiliary bus the reduction in hardware that justifies a bus technique is not realized since the auxiliary bus is really a dedicated point-to-point transfer medium.

In summary, the 1553 bus is considered adequate for current and projected needs for use as a status and control distribution medium where a high degree of fault detection capability and/or redundant distribution is required. *The 1553 bus is considered inadequate where asynchronous data transfers between avionics elements are required when the actual utilization practices are considered.*

5.3.1.3 Future Requirements

As discussed in the previous paragraphs, the MIL-STD-1553 time multiplex bus is not considered adequate for data transfers between elements particularly for asynchronous transfers. In order to correct this deficiency in a manner compatible with protecting the Navy's investment in hardware, as well as the commonality already achieved by the MIL-STD-1553 bus, the development of an augmented bus to supplement the data transfer capability of the 1553 bus is recommended. The augmented bus recommended (see 6.4.1.1) would operate in parallel with the MIL-STD-1553 bus and all data transfers on the augmented bus would be controlled by the 1553 bus. The augmented bus implementation is recommended only for new elements (remote terminals) in previously equipped MIL-STD-1553 platforms which require moderate to large data transfers. The augmented bus will require additional hardware, however, since no protocol or control is involved the hardware impact would be significantly less than the implementation of an auxiliary MIL-STD-1553 bus to perform the same function. The augmented bus should be considered an interim measure, (i.e., a simple augmenting data transfer medium).

In order to allow for more complex digital processing in future weapon systems, developments separate from the MIL-STD-1553 are also recommended. Currently a nonairborne optical fiber bus development is underway as part of the Marine Corps Tactical Air Operations Central-1985 program. There are technology investigations underway for MIL-STD-1553 optical busses as well as for higher speed electrical busses as part of the SAE-A2K committee efforts. For this reason, the requirements for a complete replacement of the MIL-STD-1553 bus were not addressed as part of this study. It is recommended that these developments continue but with some modification. The implementation of optical busses is not dependent on protocol hence optical bus developments should be dedicated to the development of optical bus couplers and interfaces independent of protocol. Additionally, the use of optical busses to provide simultaneous frequency and time multiplex capability (i.e., more than one time

multiplex bus operating simultaneously on one cable) should be investigated. The purposes of higher speed electrical busses are to increase bus efficiency and flexibility through protocol. The electrical bus developments should be dedicated to the development of protocol independent of the transmission medium. Program schedules should be coordinated so that each will benefit from the other. Additionally, parallel studies, as opposed to developments, should be conducted to determine if VHSIC technology will have impact on time multiplex bus capabilities should the optical bus hardware developments be unsuccessful.

The continuation of development efforts to provide a high speed replacement for the MIL-STD-1553 serial bus is recommended even though the parallel augmentation bus will satisfy the near term core avionics requirements. This recommendation is made primarily because the number of units that can be attached to the electrical parallel augmentation bus recommended in 6.4.1.1 will be limited (6-10 units) by the line drivers recommended in 6.2.1.2.2. The configuration shown in Figures 1 and 2 assumes 7 augmented bus connections (not including mission avionics). The configuration shown in Figure 3 assumes 10 augmented bus connections without considering mission avionics. When mission avionics and core avionics growth potential are considered, the recommended augmentation bus must be considered an interim solution for asynchronous data transfers if limited to the interface of a maximum of 10 units. An alternative to the recommended electrical interface is the use of a fiber optic link to provide augmentation of the MIL-STD-1553 bus. The fiber optic link would require only a single cable since serial rather than parallel bit transfers can be accommodated within the bandwidth of an optical bus. The implementation of an optical augmentation bus could satisfy future digital data transfer requirements without replacing the existing MIL-STD-1553 bus, however, a bus concept that includes the features of the augmented bus (higher speed as well as variable rate data transfers) and that also addresses the higher level protocol requirements for processor to processor interactions is the preferred approach.

5.3.2 Frequency Multiplexing

The use of frequency multiplexing techniques to distribute information over coaxial cables can offer many of the same advantages as time multiplexing, namely, point to point (or point to several points) reconfigurable interfaces that provide flexibility in terms of changes or additions. Frequency multiplexing offers an advantage over time multiplexing in that it is compatible with all signal types and simultaneously provides continuous outputs to all elements. The use of frequency multiplexed signals over coaxial cables is widespread in cable television systems, however, there are a number of constraints which affect the utilization of frequency multiplexing for transmission of signals in aircraft.

5.3.2.1 Bus Configurations

An ideal frequency multiplex bus for avionics equipment would allow direct exchange of information between any of the elements on the bus regardless of position on the bus. Frequency multiplex systems operate over a broad bandwidth in order to accommodate multiple channels with reasonable information content and it is the broadband operation that in general requires a directional bus. This is true for two reasons:

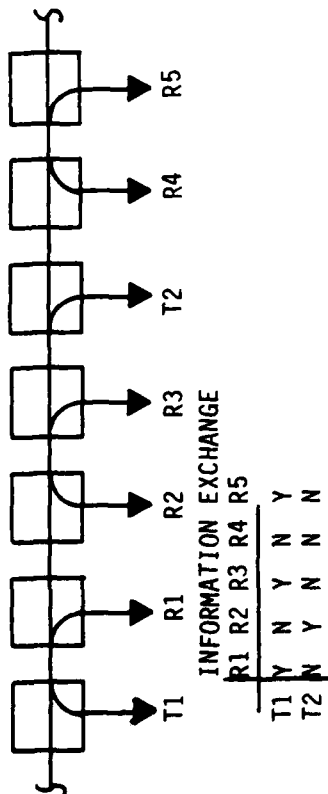
- a. In order to maintain broadband transmission characteristics directional couplers are used to inject into, and extract signals from, the bus, and
- b. to overcome the losses due to the couplers and the lossy transmission line, amplifiers which are non reciprocal devices are utilized.

Directional bus operation can provide significant flexibility in terms of reconfiguration of interfaces but the overall capability is limited by the position of the directional couplers on the bus (see Figure 4a). For example, a

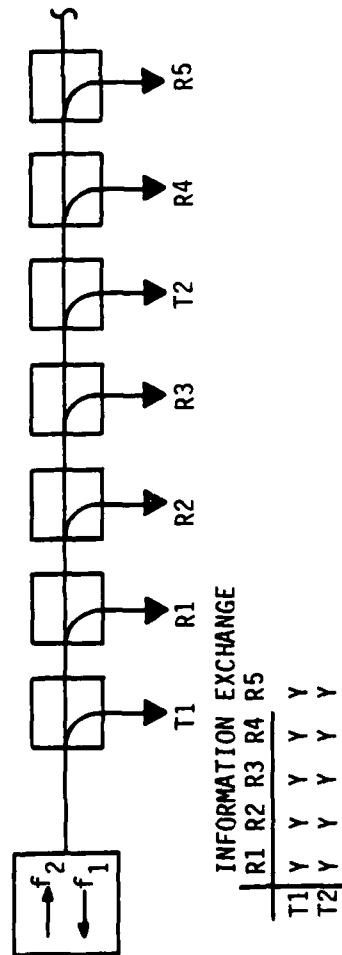
transmitter can send information to any number of receivers downstream but a receiver upstream from the transmitter (or a receiver downstream of the transmitter configured to receive signals from other transmitters further downstream) cannot receive the signal. Thus the ability to exchange information is limited by the initial configuration of the bus. The use of in line amplifiers in the position dependent configuration is limited and once incorporated further reduces bus flexibility. Commercial systems (See Figure 4b) overcome the configuration limitations by having all the transmitters send signals upstream to the "head end" where they are converted to a different channel and re-transmitted downstream for receipt by downstream receive couplers. In the cable television systems the requirements for in-line signal amplification are met by frequency dependant bidirectional amplifiers (low and high pass amplifiers in opposite directions). Thus the bus is in reality two busses operating in different directions and frequency bands on one cable with centralized control. Neither the position dependant nor the "head end" bus configurations is applicable to future avionics installations.

The position dependent configuration offers little growth capability and may require excessive cabling to attach the elements to their correct locations on the bus. The "head end" approach, although compatible with additional processing (e.g., symbol overlay on raster scan display video), requires an excessive hardware investment. For each exchange of information between two elements a receiver and transmitter at the "head end" is required along with the excess bandwidth to accommodate the two channels required for each exchange.

The directional couplers used for frequency multiplex busses are four port devices and are capable of coupling in either direction if the appropriate input/output port is selected with the other port terminated. Additionally, dual direction coupler configuration capable of injecting or extracting signals on the bus in both directions simultaneously are realizable (however for the same through insertion loss (direct path) the injection or extraction loss is approximately 6db greater than for a single directional coupler). Bus



a) Position Dependent Configuration



b) Head End Configuration

LEGEND Y = YES
N = NO

Figure 4 Directional Frequency Multiplex Busses

designs which provide dual directional capability on one cable using either coupling technique are feasible as long as the length of the bus and the number of couplers is constrained to allow operation without in-line amplifiers. Using the switched directional coupler ports, the simultaneous transmission of signals to both up and down stream receivers is not possible. Using the non-directional coupler approach severely limits the size of the bus when reasonable dynamic ranges are utilized due to the additional 6dB injection and 6dB extraction losses. Although they are practical, bus designs of this nature are not applicable to general avionics due to the inherent growth limitations.

The growth limitations of single cable dual direction busses can be overcome to some extent by the inclusion of frequency dependent bidirectional amplifiers (as in cable television) and management of frequency assignments based on the required direction of information transfers. Alternately, two busses without in-line amplifiers can be connected end to end through a low pass filter (bus length at lower frequencies may be extended due to cable loss characteristics) with low frequency assignments used for inter bus exchanges. Again, due to limitations in growth potential as well as excessive bus management requirements neither approach is considered applicable to general avionics equipment.

No single cable frequency multiplex bus configuration with general applicability to avionics equipment has been identified. Figure 5 illustrates a two cable configuration that is applicable. As illustrated in the figure, with a two cable configuration any transmitter can send information to any receiver. As noted in the figure but not recommended, receivers for which all transmitters are upstream require only a single directional coupler. The inverse is also true for transmitters. In-line amplification is feasible and the location of the amplifier(s) will not limit the bus operation. The couplers depicted in Figure 5 appear complex, however, they would require the same number of components as the dual directional coupler configuration discussed above and exhibit 3dB, as opposed to 6dB, additional insertion/extraction loss for a given value through insertion loss.

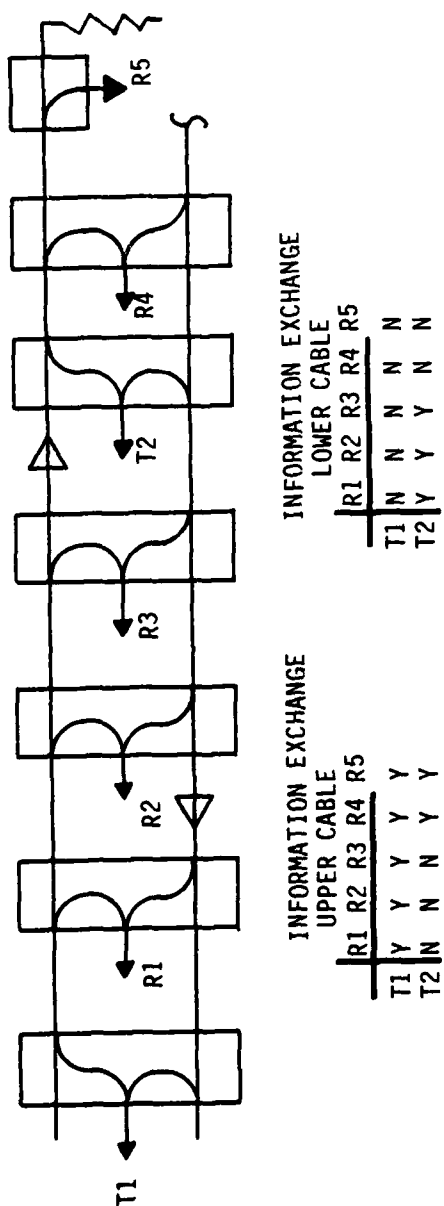


Figure 5 Two Cable Frequency Multiplex Bus

The actual insertion losses encountered by a frequency multiplex bus depend on the transmission line loss as well as the coupling values utilized. Figure 6 compares the maximum insertion losses for three bus configurations. The single cable directional coupler curve is applicable to Figure 4a. and is provided for reference only. The two cable bus coupler curve is applicable to the configuration in Figure 5 and could be reduced by 3dB with no impact on bus operation if receiver directional switching were utilized. The single cable dual direction coupler curve illustrates the additional insertion loss incurred in order to simultaneously inject a signal in two directions on one cable. The value of n represents the number of coupling devices, however, since each of the coupler types can provide two simultaneous input/output connections the total number of devices for the non switched configuration could conceivably be $2n$. The insertion loss calculations assume equal value couplers for both transmit and receive applications and a through insertion loss for each coupling device of .45 dB (actual through insertion loss equals coupling loss plus insertion loss). For each value of n the optimum coupling factor for each of the 3 configurations to minimize the worst case insertion loss was determined and the resulting worst case loss is plotted in Figure 6.

Since the total loss difference between the two cable bus coupler and the single cable directional coupler configurations can easily be overcome by in-line amplifiers, the two cable bus coupler configuration is considered the preferred configuration. The two cable configuration allows direct communication between any two elements on the bus, is independent of coupler placement, is compatible with in line amplification and thus expansion, requires no more bandwidth than is actually necessary for the transfer of information, and requires only management of frequency assignments (independent of direction).

5.3.2.2 Future Requirements

Frequency multiplex bus distribution is compatible with almost every signal type required in aircraft. Once installed in an aircraft, the number of potential users for a frequency multiplex bus is difficult if not impossible to forecast. The bandwidth requirements for each potential use is likewise

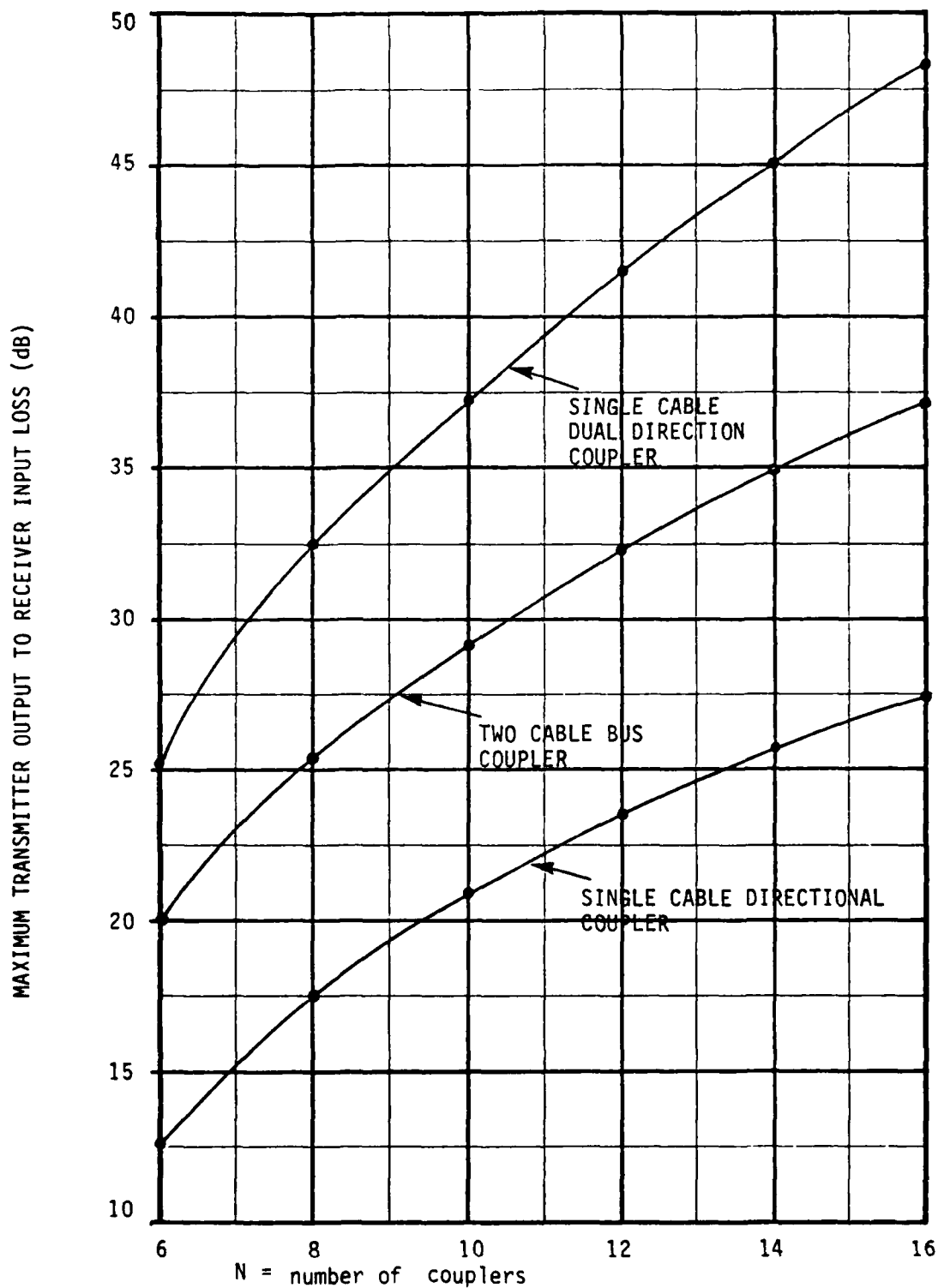


Figure 6 Maximum Coupling Loss for frequency multiplex bus optimized for the number of couplers versus the transmitter output to receiver input loss

not predictable. In order to be compatible with all standard interfaces recommended in this report, a single channel information bandwidth of 17 MHz would be required (video interface discussed in 5.1.4). Considering pulse preservation, amplitude modulation with Double Side Band (DSB) transmission is desirable, hence, a 17 MHz information bandwidth would require 34 MHz of bus bandwidth (22 MHz of bus bandwidth for vestigial sideband of sufficient quality for television sync and video). In order to simplify receiver transmitter design, the 34 MHz bus bandwidth is recommended for pulse as well as sync and video information. Operation of the bus with each channel having a bandwidth of 34 MHz would be inefficient particularly when 40 KHz of bus bandwidth is the maximum that would be required for high quality voice.

In order to allow some optimization of bus efficiency for what is basically an undefinable requirement, the following bus implementation is recommended. The bus should allow for 3 different bus bandwidth channels namely, 34 MHz (17 MHz information using DSB), 10 MHz (5 MHz information using DSB), and 1 MHz (500 KHz information using DSB). The arbitrarily recommended number of channels for each bandwidth is seven. This would result in a bus with a total of 21 channels and would require operation in the 500 MHz to 1 GHz frequency band.

It is anticipated that such a bus would be used to satisfy a number of different interface requirements as for example, signals intended for radio frequency transmission as well as signals for cockpit display of navigation information. Such applications would have Red/Black implications, for example it would not be desirable to inadvertently transmit navigation way point information. However, it may be desirable to view simultaneously a video display in the cockpit while transmitting the same information via data link. As a potential solution to this problem it is recommended that the three different bandwidth channels be spread over the operating frequency of the bus. Equipment capable of radio frequency transmission could then be required not to implement processing capability for two of the 10 MHz and 34 MHz channels

operating at the upper frequency channels. These channels could then be utilized exclusively for processing Red signals. If further restrictions for a particular aircraft application were required low pass or band pass filters could be installed at the appropriate points along the bus to preclude transmission of Red signals. The recommended implementation would require additionally that software fail-safe modes in the frequency channel assignments, similar to the terminal address restrictions being considered for Red/Black MIL-STD-1553 applications, be implemented.

The incorporation of frequency multiplex busses in future avionics installations, for which there are redundancy requirements, deserves special consideration. The provisions for true redundancy in frequency multiplex bus applications would be similar to any other redundancy implementation, i.e., standby equipment capable of performing the function for which redundancy is required. In general, some of the equipment interfaced to the multiplex bus would be capable of performing the same functions (e.g., multi-purpose cockpit displays or video data processors), hence degraded mode capability as opposed to redundancy can be maintained with the failure of the equipment. It is envisioned that in many cases the conversion of bus information back to baseband will be performed within the equipment (e.g., in the cockpit displays), hence, a baseband converter failure would be a display failure and degraded mode capability would be available using an alternate display. In this case only the bus coupler and cabling would be required to be redundant. There are other applications where a single bus to baseband converter might be utilized for two or more equipment items, however, if redundancy were required for one function then redundant bus to baseband converters would be required and it would be just as simple to make the converter part of the equipment. It is recommended that frequency multiplex bus signal converters be made part of each piece of equipment unless a determination can be made that there are no redundancy requirements for the equipment. The part of the equipment recommendation does not necessarily mean the converter is contained within the equipment, but, rather it could be satisfied with a dedicated converter for

each piece of equipment. If the recommendation is followed then only redundant bus couplers and cabling should be required.

The two cable bus configuration recommended offers some inherent redundancy in that a loss of transmission capability in one direction will not in general affect the transmission in the opposite direction, however, this feature will not permit true redundant operation. It is recommended that redundancy be achieved by the installation of two separate two wire busses. All equipment utilizing frequency multiplexing should interconnect to one bus. All mission essential equipment should be switch connectable to the redundant bus. Simultaneous coupling to both busses with almost no insertion loss impact is feasible, however, due to interference on both busses that can be generated by coupler isolated port leakage and the dependence of coupler performance on matched terminations on both busses, this technique is not recommended. Where redundant bus connection is required, two separate bus cables should be brought to the equipment interface and bus switching performed internal to the equipment.

Normal bus operation should allow utilization of the redundant bus to expand overall capability. This can be accomplished by allowing the sum of mission essential and non mission essential bus utilization to exceed bus capacity so long as the total mission essential requirement is less than the bus capacity. For example, assume the mission essential equipment required 70% of bus capacity and the non mission essential equipment required 40% of bus capacity. Further, during normal operation assume the main bus carries half the mission essential and all the non mission essential signals ($1/2 \ 70\% + 40\% = 75\%$ of bus capacity) and the redundant bus carried the other half of the mission essential signals ($1/2 \ 70\% = 35\%$ of bus capacity). A failure on the main bus would require switchover so the redundant bus carried all of the mission essential signals. This would result in total loss of the non mission essential signals. A failure on the redundant bus would require deletion of non mission essential signals in order to carry the necessary signals. The effect

of either failure would be the same i.e., deletion of non mission essential functions. Allowing this mode of operation maximizes hardware utilization.

Control of the frequency multiplex bus should be accomplished via the MIL-STD-1553 bus for a number of reasons. The serial data bus will provide redundant control as well as a means of controlling Red/Black frequency assignments. The 1553 bus will also provide an available link between different subsystems or groups of subsystems. The control function should be performed by an aircraft computer rather than within any one subsystem in order to allow integration of different equipment in different aircraft.

6.0 RECOMMENDATIONS

Specific recommendations for actions or policies that can be implemented to achieve the ultimate goals of the AVCS program are presented in this section. The recommendations given are based on considerations of all the areas discussed in this report. Negative comments or deficiencies noted in other sections of this report are countered with positive recommendations and a reference to sections of this report where italicized comments were made. Additionally, recommendations made in other portions of this report are repeated to provide a consolidated list and indicate a method of implementing the recommendation. Specific recommendations regarding AIDS, IISA, TIES, and mission equipment are not repeated. The reader is referred to 4.3.2, 4.3.3, 4.3.4, and 4.3.5 respectively for recommendations regarding these elements.

6.1 MODIFICATIONS TO EQUIPMENT PROCUREMENT PRACTICES

6.1.1 General

The implementation of individual recommendations for revised military standards, considerations to be applied to ongoing efforts, and new developments do not in general present procurement problems. The procurement of new equipment compatible with the architectural philosophy of Section 4.0 presents

both political and procurement problems. The architecture presented in Section 4.0 was generated from a system engineering point of view with the system being the core avionics. This approach leads (correctly) to functional allocations for pieces of equipment and not to requirements for individual subsystems (i.e., the equipment no longer represents a stand alone system or whole). For example, all crew interfaces for GFE core avionics are provided via the Display and Control subsystem, thus, the radio frequency control is not part of the radio. In order to procure a radio it must be tested and in order to test the equipment a frequency control is required. The ramifications of this problem are more severe for other "subsystems". The procurement of GFE inertial sensors and limited capability Navigation Processors that don't tell you "how to get home" (waypoint and dead reckoning computations done in the CFE Mission Computer) makes the equipment appear less useful (even though it may actually be more so) than an inertial navigation "system". The overall performance of the core avionics system as well as the individual equipment items procured as GFE cannot be evaluated until the integration with CFE hardware and software is accomplished (e.g., navigation accuracy is a function of the quality of the sensor and Navigation Processor but is also a function of the quality of the software in the Mission Computer which may be supplied by a different contractor).

The "who's at fault if it doesn't work" problem is not new and is not tied to the architecture of Section 4.0. If the benefits of common avionics are to be realized then common equipment must satisfy the least common denominator for all the platforms. The procurement of this equipment is manageable if performance requirements are stated relative to interface information quality and if detailed interface and control requirements are established. The demonstration of equipment can be accommodated by simultaneous development of support equipment with dynamic test capability and, if necessary, brassboard quality controls and displays for flight demonstrations. The generation of procurement packages of sufficient detail will place a larger system engineering burden and responsibility on the government but represents a cost that can yield significant benefits. Of equal importance is self restraint on the part

of the government to limit the requirements to those of the least common denominator without designing the entire system and thus stifling the ability of the airframe contractor to meet the weapons system requirements using GFE.

The interface detail required for equipment adapted via an AIU module should be stated relative to the output of the equipment developers side of the AIU module (see 6.4.2.1.3) with no detail requirements for the AIU to equipment interface other than a description of the information to be accessed from the AIU. For example, if it is desired to tune a radio via a serial digital bit stream then that is the requirement that should be stated for the AIU. If the contractor chooses to convert this to parallel data within his half of the AIU he should be free to do so. This approach is consistent with the least common denominator in that another aircraft that may require parallel data can be accommodated with only a different AIU conversion. It is also consistent with minimizing the excess hardware within the subsystem since, any interface specified for the AIU to equipment connection would most likely require some form of conversion within the equipment to accommodate the equipment mechanization (e.g., fine and coarse tune signals). Equipment intended for interface via an AIU should state the interface method explicitly. Testing of AIU interfaced equipment should be accommodated using prototype half AIU modules with less than half of the module capacity utilized in order to allow for aircraft peculiar requirements.

Equipment that has the potential for multipurpose use (e.g., radio transmitter/receiver assets) should be required to be packaged as separate units (deficiency noted in 5.2). Additionally, multipurpose equipment specifications should be reviewed for TIES compatibility prior to initial development.

The equipment specification should detail the requirements for multiplatform application. Consider the time multiplex bus address programming for each piece of equipment via the aircraft interconnect that is recommended in 6.2.3.1.1. This implementation requires that provisions must be made within the equipment for variable address assignment for associated equipment. For

example, if it is required that the two Display and Control Processors communicate with each other over the mission computer time multiplex bus then provision must be made within each processor to accept and store the address of the other processor from the Mission Computer as part of the power up sequence.

In order to achieve the maximum benefit from common equipment the development of a single piece of equipment must consider the capabilities of other equipment as available assets. For example, for built-in test purposes an on-board radiated interference signal is a BIT oscillator if the blanking signal is disabled during the test. Such implementations are possible for RF checks of IFF transponder/interrogator, IFF/TACAN, Radio/ADF/DDI, and EW/Radar Beacon/Beacon Augmenter/ILS if signal acceptance criteria in BIT were specified to be compatible.

6.1.2 Functional Interfaces

It is recommended that the practice of developing replacement equipment which is a "form, fit, functional replacement" for existing equipment be discontinued particularly in regard to interfaces. Producing a subsystem that has, for example, synchro outputs to accommodate older aircraft and a digital output for newer aircraft means that every aircraft requires increased subsystem acquisition costs; increased size, weight, power consumption, cooling; and decreased reliability (more parts) as well as the potential for inconsequential failure detections (faulty synchro output detected in a digital aircraft). Although it is politically attractive to advertise a development equipment as a direct replacement, it almost never actually is. New equipment provides new capability which most often requires aircraft integration to allow its use. Equipment that provides the lowest common denominator in capability is the most likely to receive universal use. When an additional interface must be provided (e.g., synchro output) this interface should be provided by a separate element (e.g. AIU) which can be GFE or CFE. This approach has a short term logistics burden, however, if the least common denominator

element is universally applied the long term result will be a reduced logistics burden.

Although somewhat contrary to the above it is recommended that specifications for certain functional interfaces be generated and incorporated in development specifications for all avionics equipment by reference. The functional interfaces should be required if applicable, regardless of whether they are needed for operation of the subsystem under development. These interfaces are required to allow a certain degree of commonality for aircraft interconnect requirements as well as for greater functional commonality.

6.1.2.1 Blanking

For electromagnetic compatibility and general integration purposes, all subsystems that transmit or receive energy in the frequency range of 30 MHz to 32 GHz should be required to incorporate blanking interfaces. The recommended interface is a ten bit code with each bit representing an octave or a suboctave above or below 1 GHz. The lower two frequency ranges should be rounded off to 30-60 MHz and 60-125 MHz. Each subsystem need only generate or accept the applicable bits of the code and should utilize the standard dedicated digital interface of 6.2.1.2.1. The blanking interval should be designated as a binary 1 or Mark state to accommodate fail safe provisions. For pulse modulated transmitters, the blanking interval should be present during the entire transmitted pulse plus 200 nsec to allow for transmission line ringing. The specification should require the blanking signal to lead the actual RF energy for transmitter outputs in order to accommodate distribution delays for the blanking signal. Subsystems requiring higher frequency resolution than provided by the specified blanking inputs should generate and accept the specified inputs in addition to any finer resolution signals. This requirement is considered necessary in order to accommodate other subsystems that cannot provide the higher resolution. The line driver for each blanking output should

be monitored internal to the subsystem to insure that artificial blanking signals due to a line driver failure, which may disable another subsystem, cause all blanking outputs for that line driver to be removed.

6.1.2.2 Clock

Each subsystem that accepts or generates digital information other than binary state information should be required to accept a square wave master clock input. A clock frequency of 250 KHz (4 μ sec Pulse Repetition Period (PRP) or 500 kilobaud) is recommended in 6.4.2.4. All digital interfaces should be required to operate at frequencies that are simple derivatives of the clock frequency. The subsystems should be required to operate with or without the clock input, however, when the clock signal is present the digital interface operation should be synchronized to the master clock within 12.5% of the nominal PRP of the interface. When the clock input is not present the subsystems should be capable of maintaining internal clock operation with synchronization provided by MIL-STD-1553 serial bus Synchronize mode commands. The imposition of these requirements will allow greater future integration by providing a mechanism for skew compensation, allowing monitoring of existing interfaces by future subsystems, and providing standard operating rates.

6.1.2.3 External Control

In order to achieve functional commonality subsystems must be capable of performing more than one operation even though they are built to perform only one function. Capability to perform more than one function can be gained by requiring access to subsystem assets. Two areas where this capability is realizable by specifying the requirement in the procurement of the subsystem have been identified.

6.1.2.3.1 Amplitude Modulation

All subsystems capable of transmitting or receiving pulse modulated RF signals should be required to provide access to the transmitter receiver assets. For subsystems with transmitter capabilities, a command override input to the subsystem should allow external modulation of the transmitter using either the video interface recommended in 6.2.1.4 for transmitters with linear capabilities or using the digital interface recommended in 6.2.1.2.2 for transmitters with fixed power output. For subsystems with receiver capabilities a single, parallel output of detected video interfaced using either the video output recommended in 6.2.1.4 for receivers with proportional amplitude characteristics or the digital interface recommended in 6.2.1.2.2 for receivers with pulse present capabilities should be required. Additional transmitter or receiver capabilities built into the subsystem to accommodate general purpose interfaces are not recommended unless they can be tied to a specific requirement. As an example of the capability that could result from the use of such interfaces, if the amplitude modulation interfaces were included along with some frequency control interfaces in modern Deceptive Electronic Countermeasures (DECM) Subsystems (e.g., AN/ALQ-126B and AN/ALQ-165) the transmitter and receiver hardware for radar beacon equipment (e.g., AN/APN-154 or AN/APN-202) could be eliminated. The example given is impractical only because DECM equipment is not carried on all aircraft or even in every flight of provisioned aircraft.

6.1.2.3.2 IF Access

All radio communications equipment utilizing more than one IF conversion should be required to provide external access to the IF for both receive and transmit capability. This interface would allow external equipment to utilize the transmitter and receiver assets. The first IF should be required to be compatible with the recommended interface in 6.2.1.5 and will then also be compatible with the multiplex switch network recommended 6.4.2.3. Access to a

single IF interface for both transmit and receive should be provided by a command override receive or transmit input to the subsystem.

The use of both the IF access and IF multiplex switching allows sharing of both the "front" and "back" end of radio equipment (see Figure 2) in order to reduce the proliferation of equipment, enable the sharing of assets to maintain capability in the presence of a failure, and accommodate future requirements by replacing only one end of the equipment. If available in existing equipment today, this interface would allow elimination of the auxiliary UHF radio receiver by utilization of the redundant radio assets while providing greater transmit/receive redundancy capability than is currently available even with the extra radio.

6.2 STANDARDS CHANGES/ADDITIONS

In order to achieve commonality, growth capability, and technology transparency military standards must be applied to avionics developments (deficiency noted in 5.1). The success of this approach is best exemplified by MIL-STD-1553. Once established, standard application must be enforced vigorously, however, provisions must be made to accommodate the legitimate needs of specialized subsystems. The application and enforcement of Qualified Parts Lists (QPL) and the Standard Electronic Module Program (SEMP) have significant effects on logistics support, however, they do not assure that avionics architectural compatibility requirements will be met in today's changing technology environment. Subsystems under development require the use of nonstandard parts in order to meet both performance and packaging requirements. There is no reason to believe this trend will not continue or be applicable to interface components. The use of detailed standards is not a cure-all for commonality but standards can effect compatibility with a large degree of technology transparency. The detail of the standards should be sufficient to insure compatibility between designs built to the same standard but no more restrictive than is absolutely necessary. Once developed, it is recommended the NAVAIR

immediately and unilaterally adopt the recommended standards in all cases where conflicts with existing military/international standards do not exist. In order to make the conversion to standard interfaces effective in terms of future aircraft integration capabilities, it is further recommended that these standards be applied to all Navy avionics developments including those of the AVCS program. The following recommendations for standards are made.

6.2.1 Dedicated Interfaces

It is recommended that a military standard for dedicated interfaces be generated. Additional standards for each specific type of interface should be generated as appendices or as dash amendments to the basic standard (as with MIL-STD-188). The quantity of additional standards should be restricted by generating standards only for a general type of interface. If the quantity of standards is not restricted, the standards cannot accomplish the purpose of compatibility between subsystems.

6.2.1.1 Binary Interfaces

Standards for only three types of binary interfaces are recommended.

6.2.1.1.1 Power Input

One on/off interface per subsystem should be allowed. The interface should operate from a switched ground return to 28 V dc (maximum current 0.2 ampere) as discussed in 5.1.1.1.

6.2.1.1.2 Current Sink Output

High impedance/low impedance outputs capable of sinking 0.2 ampere from 28 V dc should be allowed for cockpit/station warning alarms. Subsystems should provide only continuous outputs with blinking applied in the instrument panel.

6.2.1.1.3 Discrete Input/Output

All other binary interfaces should be identical to one of the two digital interfaces recommended on 6.2.1.2.

6.2.1.2 Digital Interfaces

Standards for only two types of dedicated digital interfaces are recommended at this time.

6.2.1.2.1 Line Driver/Receiver

It is recommended that a general military standard applicable to all Navy avionics equipment that adopts EIA-RS-422A without modification be generated (deficiency noted in 5.1.2.1). The standard should require line drivers to operate at up to a 10 MHz rate with multiple line receivers spaced anywhere from 0 to 150 feet from the driver with all line receivers utilizing input terminations for ac impedance matching and with or without internal fail safe provisions incorporated (deficiency noted in 5.1.2.1). This requirement could be stated in terms of compatibility with a specific receiver chip (e.g., MIL-M-38510/104A-04) and specified terminations and if necessary should reduce the number of parallel receivers from 10 as required by EIA-RS-422A to a minimum of six (deficiency noted in 5.1.2.1). The potential of increasing the number of line receivers beyond 10 (limit imposed by EIA-RS-422) by utilizing single point, fail safe biasing (pull up-pull down resistors) should be investigated and is recommended as part of the augmented bus development in 6.4.1.1. In order to allow for future integration the standard should require initial designs to be configured such that no more than 70% of the maximum number of parallel line receivers is driven by a single line driver. Line receiver requirements should be stated in terms of compatibility with the maximally loaded line driver.

6.2.1.2.2 Tri-state Line Driver

It is recommended that a general military standard, applicable to all Navy avionics equipment for a tri-state line driver be generated. The line driver should be a one for one replacement for the line driver of 6.2.1.2.1 (e.g., MIL-M-38510/104A-05) and be compatible with all requirements for the line receiver loads of 6.2.1.2.1. For transceiver operation (half duplex), the interface should allow "wire or" of the tri-state line driver and line receiver external to the equipment.

Utilization of the tri-state line driver should be required in lieu of the driver of 6.2.1.2.1 in all cases where the information supplied by the line driver can be utilized in response to an external input from another subsystem. This implementation mechanism is directly compatible with bus requirements and offers growth capability when non-bus interfaces are employed. For non-bus interfaces a tri-state line driver makes the line receiver end of the interface compatible with multipurpose uses.

6.2.1.3 Analog Interfaces

Standards for only two types of analog interfaces are recommended at this time.

6.2.1.3.1 Amplitude

A general military standard applicable to all Navy avionics elements for proportional amplitude interfaces should be established. The standard should allow one unipolar type (0 to +10 V recommended) and one bipolar type (-5 to +5 V recommended), amplitude proportional interface which is compatible with analog/digital sampling or digital/analog conversion.

6.2.1.3.2 Phase

A general military standard applicable to all Navy avionics elements for proportional phase interfaces should be established. The standard should allow for one low level type (26 V (11.8 V signal)) and one high level type (115 V (90 V signal)) phase proportional interface which is compatible with synchro/digital sampling or digital to synchro conversion.

6.2.1.4 Video Interfaces

It is recommended that a general military standard applicable to all Navy avionics equipment be developed for a 17 MHz bandwidth video interface. The interface should accommodate both pulse video (120 nsec pulse width minimum and 40 nsec rise time maximum) and composite sync and video raster scan signals (actual requirement $\approx$ 16.9 MHz for 1023 line (with equal vertical and horizontal resolution) 4:3 aspect ratio sync and video display). Group delay requirements should also be established to accommodate multichannel pulse comparison systems.

6.2.1.5 Modulated Carriers

As discussed in 5.1.5 general standards for modulated carrier interfaces are not recommended because of potential subsystem performance considerations. A military standard applicable to all Naval avionic radio communications equipment for IF interfaces is recommended. The recommended interface is a 70 MHz IF with a 10 MHz bandwidth at a nominal -20dBm signal level.

6.2.2 Switched Interfaces

Because of the specialized nature of switched interfaces as discussed in 5.2 it is recommended that standards for switched interfaces not be adopted. However, as part of the overall interface standard it should be required that all switching functions be performed by a standard interface and that switching networks be used only to switch standard interface signals.

6.2.3 Multiplexed Interfaces

6.2.3.1 Time Multiplex

In addition to the following standard revisions the development of an augmented MIL-STD-1553 data bus and an associated standard is recommended in 6.4.1.1.

6.2.3.1.1 Recommended Revisions to MIL-STD-1553

- a. The maximum number of data words per transfer should remain at 32 for serial bus transfers but allowance should be made to incorporate larger transfers via a parallel bus as indicated under the recommended developments (deficiency noted in 5.3.1.2).
- b. The standard should be revised to require that terminal addresses be programmable by grounding of five pins by aircraft wiring in a manner consistent with the AN/ARC-182 convention (deficiency noted in 5.3.1.2).
- c. The standard should be revised to require that both long and short stub interconnects be provided for each element. This is necessary to allow flexibility of installation in different weapon systems.
- d. The subaddress field of the Command Word should be specified in order to gain a larger degree of commonality between weapon system applications. The specification of the subaddress field should be accomplished considering all known applications to provide maximum compatibility with existing uses. The latitude required for system design can be achieved using single data word transfers (16 bits of control) along with the Command Word.

- e. The specification of the subaddress field or an addition to the mode command field should allow for: transmission of one and two data words in response to a vector word mode command, transmission and reception over one or more alternate busses (parallel bus for example), and transmit new status word to allow for service requests as part of a normal polling cycle.

6.2.3.2 Frequency Multiplex

The establishment of a military standard for signal distribution by frequency multiplex is recommended. This standard should be generated during the interface development recommended in 6.4.1.2.

6.3 CONSIDERATIONS FOR ONGOING/PLANNED DEVELOPMENTS

6.3.1 Intercommunications System (ICS)

The ICS subsystem being developed under the AVCS program should include the following: personnel equipment with redundant speakers and microphones that meet TEMPEST requirements, redundant Communications Security (COMSEC) switching units, and self-compensating (one volume control for all functions) redundant audio switching and amplification units. This subsystem should be designed to be modularly expandable to accommodate one or more crew members. Operator controls, including COMSEC switching, should be redundant and self contained in the ICS subsystem. General audio summing should not be part of the system. The system should allow for dual inputs/outputs of audio for each redundant path and allow for a single combined tone input in each redundant path.

6.4 NEW DEVELOPMENTS

6.4.1 Interfaces

6.4.1.1 Augmented MIL-STD-1553 Bus

The utilization of serial data busses configured according to MIL-STD-1553 is increasing and represents a substantial investment on the part of the Navy. This bus system has several limitations in terms of its general application as a digital data distribution medium (deficiency noted in 5.3.1.2). The continued proliferation of 1553 data busses will yield; multiple bus configurations with multiple bus controllers to overcome the shortfalls of a single (redundant or not) main bus, and bus utilization fine tuned to the initial system design thus prohibiting the use of any inherent growth capability (i.e., MIL-STD-1553 compliance in the letter and not the intent). The Navy can protect the existing investment in 1553 bus hardware while providing additional flexibility to the system designer and avoiding fine tuned systems by augmenting the capability of the 1553 bus.

The recommended development consists of adding an augmenting bus to the MIL-STD-1553 configuration. The augmenting bus would have no protocol, no bus controller, and no control capability. The use of the augmenting bus would be entirely under the control of the serial data bus controller. The augmenting bus would allow high data rate transfers of Block and String information between any elements of the system without restricting the use of the serial bus. The processing of requests for augmenting bus utilization and the assignment of the transmitting element and the receiving elements for the augmenting bus would be through and by the standard 1553 serial bus.

The implementation details require study beyond the scope of this effort, however, to illustrate the concept the following details are given as a possible means of implementation. A representative configuration is shown in Figure 7. An eight bit plus parity parallel bus augmenting configuration is defined. Data format would be NRZ (nonreturn to zero) at a 500 KHz rate. Each bit and the parity bit would be transmitted via a balanced voltage interface circuit which would be interfaced as a tri-state buffer (recommended in 6.2.1.2.2). This interface would limit the number of units connected to the

bus to approximately 10, however, the use of bus biasing provisions in only one unit to allow a greater number of units should be investigated. If the number of units connected to a single electrical augmented bus cannot be increased beyond 10, bidirectional buffers, with a one bit time delay to extend the bus, would be feasible and could be incorporated in the Mission Computer(s); however, this approach is not recommended since it restricts the flexibility in bus implementation.

Using the configuration of Figure 7 and the same conditions discussed in 5.3.1.2 (for the standard 1553 remote terminal to remote terminal asynchronous data transfer) yields the following capabilities. Using 160, 80, 40, and 20 msec bus controller cycle times and capability for 30 terminals results in 5333, 2666, 1333, or 666 μ sec of bus utilization time to service each remote terminal. In order to set up a remote terminal to remote terminal transfer, approximately six Command Words, five Status Words, and four Data Words over the serial bus are required. After the transfer over the parallel bus is complete, two Command Words and two Status Words are required on the serial bus to check the transfer and terminate the transaction. The utilization time of these transfers on the serial bus is between 408 and 520 μ secs. For the assumed cycle times, the size of the Block or String transfers of 16 bit words in binary word group sizes that could be transmitted over the 500 KHz parallel bus in the remaining time is (note figures in [] represent 1553 serial only values from 5.3.1.2): one 32 word transaction [0 @ 32 words] at the 50 Hz (666 μ sec/terminal) sampling rate; one 128 word transaction [1 @ 32 words] at the 25 Hz (1333 μ sec/terminal) sampling rate; one 512 word transaction [2 @ 32 words] at the 12-1/2 Hz (2666 μ sec/terminal) sampling rate; one 1024 word transaction [4 @ 32 words] at the 6-1/2 Hz (5333 μ sec/terminal) sampling rate. This results in a 3200 word/sec rate for transfer between remote terminals when operating with the 50 or 25 Hz cycle and a 6400 word/sec rate for transfer between terminals when operating with the 12-1/2 or 6-1/4 Hz cycle as compared to 800 words/sec for three of the four cycles with the standard 1553 serial remote terminal to remote terminal asynchronous transfer.

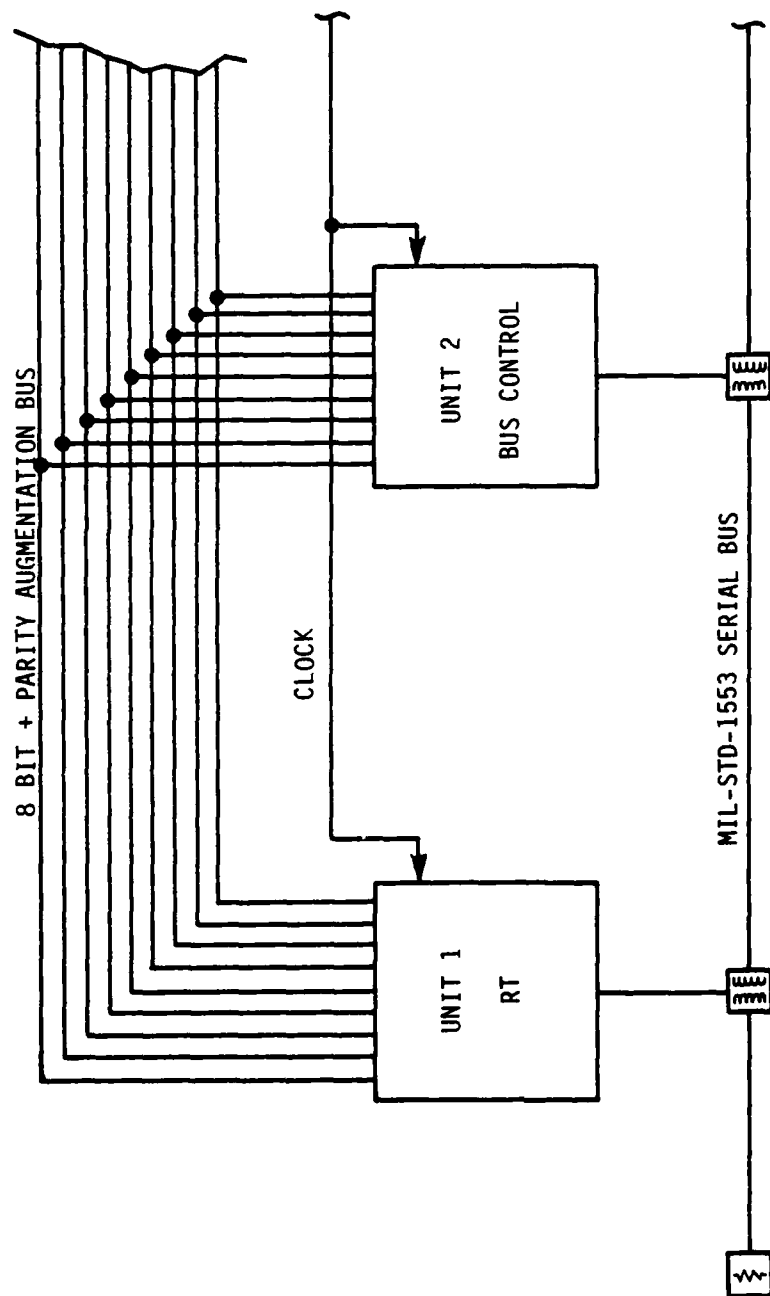


Figure 7 Augmented MIL-STD-1553 Data Bus

Despite the increased transfer capability, there are several other factors to consider regarding the advantages of an augmenting bus. The data rate on the parallel bus assumed in the above examples was 500 KHz (chosen to simplify processing of the data and synchronization); however, the electrical interfaces would be capable of up to approximately 10 MHz rates. The actual implementation of a parallel augmentation bus should consider selectable transfer rates so that an individual transfer can be optimized for the participants (example, slower rates for a load from tape and higher rates for a memory to memory transfer). The above example assumed that the parallel bus transfer must be completed within the serial bus polling cycle time allocated to one of 30 elements, however, if no other parallel bus transfers were allowed until the next complete cycle of the bus controller, the entire bus controller cycle time could be used for a parallel bus transfer (increase in transfer rate by a factor of 30) without affecting the operation of the serial bus. Additionally augmented bus transfers could be broadcast to several users without inhibiting the serial bus operation of the same users. Although not recommended, because of the inflexibility of the control requirements, the augmented bus could also be shared between elements on different serial busses.

The serial bus transfer from remote terminal to remote terminal operating at 6-1/4 Hz sampling rate would require 1.125 seconds to transfer a single 1024 word block (7 cycles at 128 words per cycle plus actual transfer time of last 128 words) whereas using the parallel augmentation bus operating at 500 KHz, the transfer would require less than 5 msec. Thus, the receiving processor could utilize Block or String type information of this size with orders of magnitude less delay for the parallel augmentation bus configuration.

Although illustrated with an electrical parallel bus, the augmenting bus approach is entirely compatible with emerging technology in optical buses where, because of data rate capability, a parallel configuration is not required. If the number of elements attached to the augmenting bus cannot be increased beyond 10 using the line drivers recommended in 6.2.1.2.2, an optical fiber implementation is the recommended approach.

The parallel interface processing would be a moderately good candidate for VHSIC technology where parity checking and word formatting could be performed in the interface chip or chips.

6.4.1.2 Bidirectional Frequency Multiplex Bus

For general distribution of interface signals in a manner compatible with the optimum utilization of common avionics equipment, the development of a bidirectional frequency multiplex bus is recommended. The recommended bus discussed in 5.3.2.2 would provide:

- a. 7 channels of 34 MHz bandwidth (17 MHz information)
- b. 7 channels of 10 MHz bandwidth (5 MHz information)
- c. 7 channels of 1 MHz bandwidth (500 KHz information)

with the different bandwidth channels intermixed over an approximate operating frequency range of 500 MHz to 1.0 GHz.

The development would be for a bus configuration as illustrated in Figure 5 and would include: bus frequency conversion modules, bus couplers (transmit, receive, and transmit/receive), and in-line amplifier and filter assemblies.

The bus should be compatible with up to 300 feet of cable and 30 couplers by the use of the in-line amplifiers. The frequency converters should, as a minimum, be compatible with:

- a. Baseband conversion for the video interface signals of 6.2.1.4 (17 MHz information channels only).
- b. 70 MHz IF conversion compatible with the interface of 6.2.1.5 (all three bandwidth channels).

- c. Baseband conversion for the amplitude analog interface of 6.2.1.3.1 (500 KHz information channels only).

However, the module for conversion from the bus frequency to the desired interface need only incorporate one interface output. The up converter for on bus coupling should be compatible with all interfaces applicable to the channel capability (i.e., convertors with 17 MHz bandwidth channel assignments should be compatible with accepting either video or IF input signals).

6.4.2 Hardware Developments

6.4.2.1 Avionics Interface Unit (AIU)

The proposed architecture (see Figures 1 to 3) is based on the utilization of standardized AIU's. These units will perform functions similar to those performed by the Computer Signal Data Converter (CSDC) in the F-14A and the Communications Systems Control (CSC) in the F/A-18. However, the proposed architecture differs from that of the F-14A or F/A-18 in several ways.

6.4.2.1.1 Concept

In the proposed architecture, a minimum of two AIU's are utilized in order to allow redundant or back-up capabilities in case of an inoperative AIU due to failure or damage. By functional partitioning between AIU's, back-up capability is maintained, e.g., TACAN and ADF separated, ACLS and DDL separated. Utilizing redundant assets via separate AIU's removes a central point system failure, e.g., two UHF radio communications systems have totally separate redundant paths. Redundant audio and tone generators as well as partially redundant display and control interfaces could allow system operation with partial failures in both AIU's. Subsystems that do not require redundancy interface with only one AIU.

The reasons for using a multiple AIU approach are several:

- a. Provide modularized I/O to accommodate existing systems as well as future technology.
- b. Reduce interconnect wiring to reduce weight and failures by incorporating interconnects within the AIU.
- c. Allow redundancy where it is required and utilize nonredundant subsystem functions within the same architecture.
- d. Allow control of subsystems within the architecture to minimize interconnects and accommodate human factors considerations.
- e. Allow for growth without drastic changes in the architecture while maintaining maximum practical commonality.
- f. Eliminate the need to simultaneously provide for multiple platform interface requirements within the individual subsystems.
- g. Separate signal conditioning from signal processing to allow for expansion of processing capabilities.
- h. Allow for aircraft peculiar requirements in order to effectively implement achievable commonality.
- i. Allow for upgrading subsystems by minimizing the impact of changes due to the utilization of individual subsystem interface modules.
- j. Provide a mechanism whereby subsystem developers can supply parts of the aircraft interface and can be held responsible for performance while allowing for Navy commonality.

6.4.2.1.2 Requirements

The general functions to be performed by the AIU's are as follows:

- a. Adapt subsystem (either existing or future) interface requirements to the avionics architecture.
- b. Provide interconnect wiring between subsystems.
- c. Replace aircraft blanking functions.
- d. Perform fail safe switchover of redundant assets.
- e. Provide single point fail safe biasing for line receivers interfaced to the AIU.
- f. Provide unique display and control command/output functions for all subsystems including warning lights.
- g. Provide redundant (between AIU's) antenna control for flight safety subsystems.
- h. Provide distribution bus adaptor ports (example, MIL-STD-1553/A/B long stub transformers).
- i. Provide distribution bus terminations where applicable.
- j. Provide device code reassignment to accommodate multiplatform subsystems.
- k. Provide buffer and distribution of master clock signals.
- l. Provide low frequency sync signal to advisory and warning displays.
- m. Provide audio volume compensation and generate all warning tones.

- n. Provide broadcast of flight dynamics to all subsystems (velocity, altitude, drift angle, pitch angle, roll angle, roll rate, and heading rate).

6.4.2.1.3 Configuration

A configuration for the AIU which will allow maximization of compliance with the avionics architecture requirements has been generated. The configuration, conceptualized in Figure 8, consists of:

- a. Government Furnished Equipment:
 - o AIU case and power distribution backplane wiring
 - o Input power filtering and DC power supply module
 - o General purpose microprocessor with random access memory
 - o Audio output and tone generator
 - o Blank modules for unused capacity.
- b. Platform Contractor Furnished Equipment (PCFE):
 - o Mission computer interface module
 - o Display and control interface module
 - o Signal distribution backplane wiring
 - o Extender module to allow piggyback, if required.
- c. Subsystem interface modules produced by the subsystem manufacturers for each platform if required. Control of the AIU interface would be via interface control agreements between the platform and subsystem contractors. A two card module is envisioned where the design of the AIU interface card is controlled by the platform contractor and built by the subsystem contractor with the other card (subsystem interface card) designed and built by the subsystem contractor.

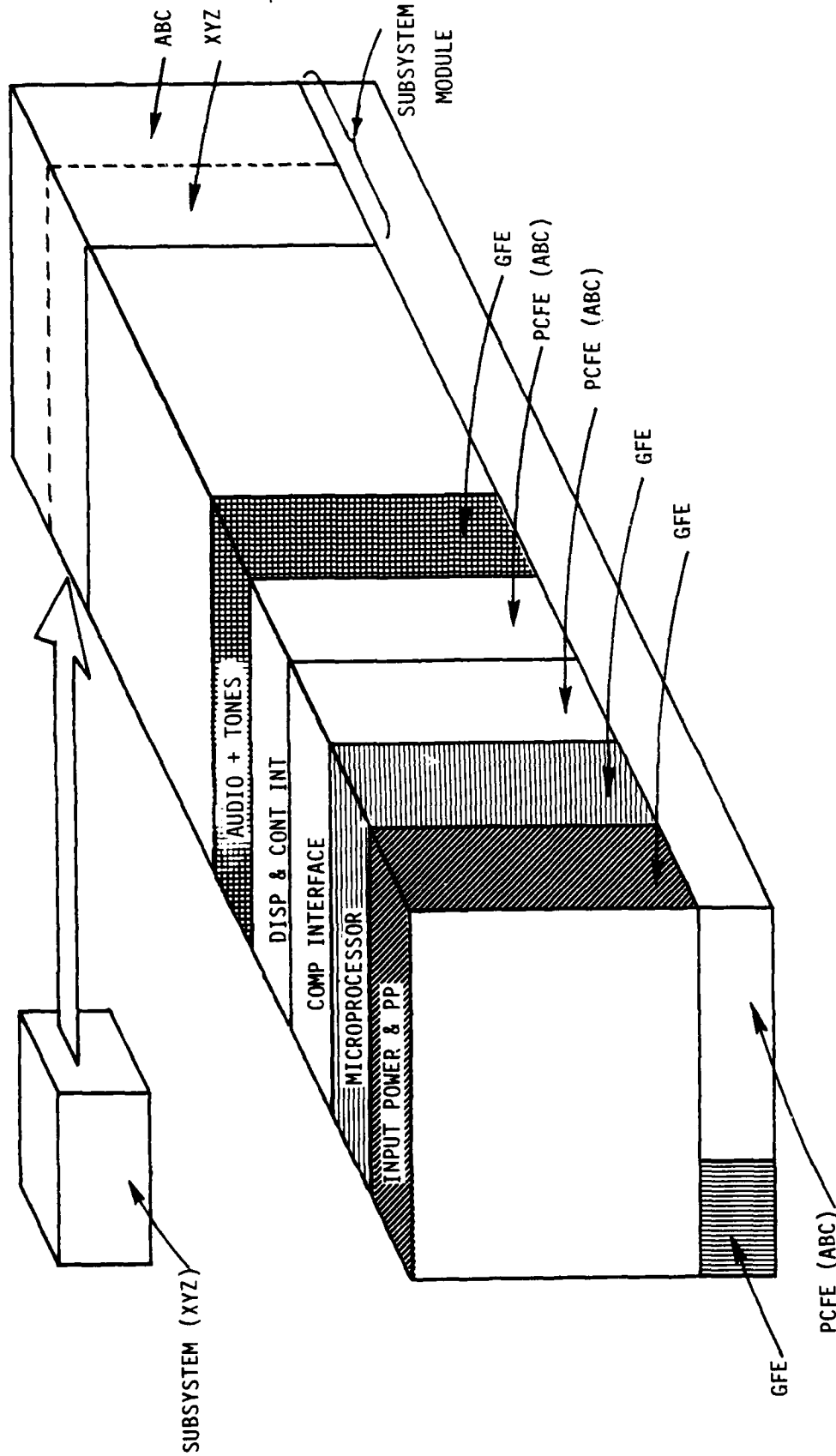


Figure 8 Avionics Interface Unit Configuration

6.4.2.2 Antenna Selection Unit

In order to minimize the impact of aircraft "antenna farms," particularly on tactical aircraft, the airframe manufacturer typically designs, builds, and tests an antenna selection unit to allow subsystems to share limited antenna "real estate." The efforts of the individual contractors should be consolidated to provide a GFE unit.

6.4.2.2.1 Concept

The antenna selection unit should be controlled via the AIU's as shown in Figures 1-3 in order to serve the requirements of the individual subsystems. Fail safe interconnects of the AIU's will be required for redundancy and back-up subsystems to preclude the possibility of an AIU failure disabling the subsystem.

6.4.2.2.2 Requirements

A low loss 30 MHz to 2.0 GHz band antenna sharing/switching unit is required to provide VHF/UHF communications, UHF DDL, TACAN, and IFF upper and lower hemisphere coverage and provide switching for the lower hemisphere UHF ADF. Configurations will require fail safe operation for: lower hemisphere VHF/UHF Communications, TACAN operation in the presence of a UHF ADF failure, UHF ADF operation in the presence of a TACAN failure, and upper hemisphere IFF operation. The configuration must allow for four VHF/UHF radio receivers and three VHF/UHF radio transmitters with fail safe operation for either of two R/T units. Additionally, manual or automatic upper lower hemisphere antenna selection (UHF, or TACAN/IFF) must be provided. Simultaneous operation of one UHF R/T, UHF ADF, and either TACAN or IFF is required.

6.4.2.2.3 Configuration

The recommended configuration for the Antenna Selection Unit is two identical fail safe RF switching units controlled by redundant AIU interface elements as configured in Figures 1-3. Antenna switching logic should be controlled by the CFE display and control module to allow for different aircraft configurations. The upper lower antenna selection circuitry should be part of the GFE audio and tones interface module.

6.4.2.3 Load and Memory Device

The architecture of Figures 1-3 assumes that a preflight Load and Memory Device will provide initial avionics control settings, a failure recording mechanism, backup storage of current configuration for both core and mission avionics, and a priori data input. The development of this device should be pursued in a manner consistent with providing a GFE unit.

6.4.2.3.1 Concept

The Load and Memory Device would be controlled entirely by the Mission Computer(s). Information storage and retrieval would be via remote terminal to remote terminal block transfers over the augmented MIL-STD-1553 bus. The main purpose of this device would be to configure variable memory prior to flight and not to provide mass storage for individual subsystem memories.

6.4.2.3.2 Requirements

The actual requirements for the intended use could most probably be satisfied by an electronically alterable, 500 kilobit memory with a 10 kilo bit/second access time. Technology in bubble memories that exists today will allow a 2 megabit memory (125,000 16 bit words) with a 100 kilobit/second (6,250 16 bit words/sec) memory to memory transfer rate. Since the technology

exists to provide for excess above the actual requirements in a small volume package the pursuit of this technology to satisfy the requirement is recommended.

6.4.2.3.3 Configuration

The configuration envisioned is for a cockpit/station mounted unit with a portable plug-in module (cassette) containing the memory device. The proposed use would require classified (confidential) cassettes, however, it may allow declassification of units that currently contain classified permanent memory (e.g., threat tables in EW equipment).

6.4.2.4 Clock

The architectures of Figures 1-3 assume an independent clock signal will be provided to elements with time multiplex digital interfaces. The clock signal would be utilized to provide simplified synchronization of the parallel portion of the augmented MIL-STD-1553 bus. Additionally, the clock would provide digital output for cockpit/station time of day display.

6.4.2.4.1 Concept

The clock element is assumed to be part of the Display and Control Subsystem with time of day setting via the Mission Computer(s) using either an external input (broadcast time reference or deck edge/microwave deck link used for inertial alignment) or manual input through a control keyboard.

6.4.2.4.2 Requirements

The recommended augmented data bus transmission rate is 500 KHz, hence, a square wave clock rate of 250 KHz would allow synchronization (500 kilobaud). The clock should operate with a 10 MHz reference to allow high speed data synchronization. The clock should provide at least five external sync outputs

via the line driver recommended in 6.2.1.2.1 in order to insure compatability with driving a minimum of 30 units without additional buffering. The clock should also be capable of synchronizing to one external input. This input could be utilized for synchronizing clocks between aircraft (via JTIDS for example) in order to accommodate cooperative tactics (cooperative jamming for example).

Redundant clock interconnections are not recommended. In the event of clock failure a backup synchronization mechanism exists via the MIL-STD-1553 serial bus Synchronize mode commands and use of the internal clock in each equipment interfaced to the augmented bus.

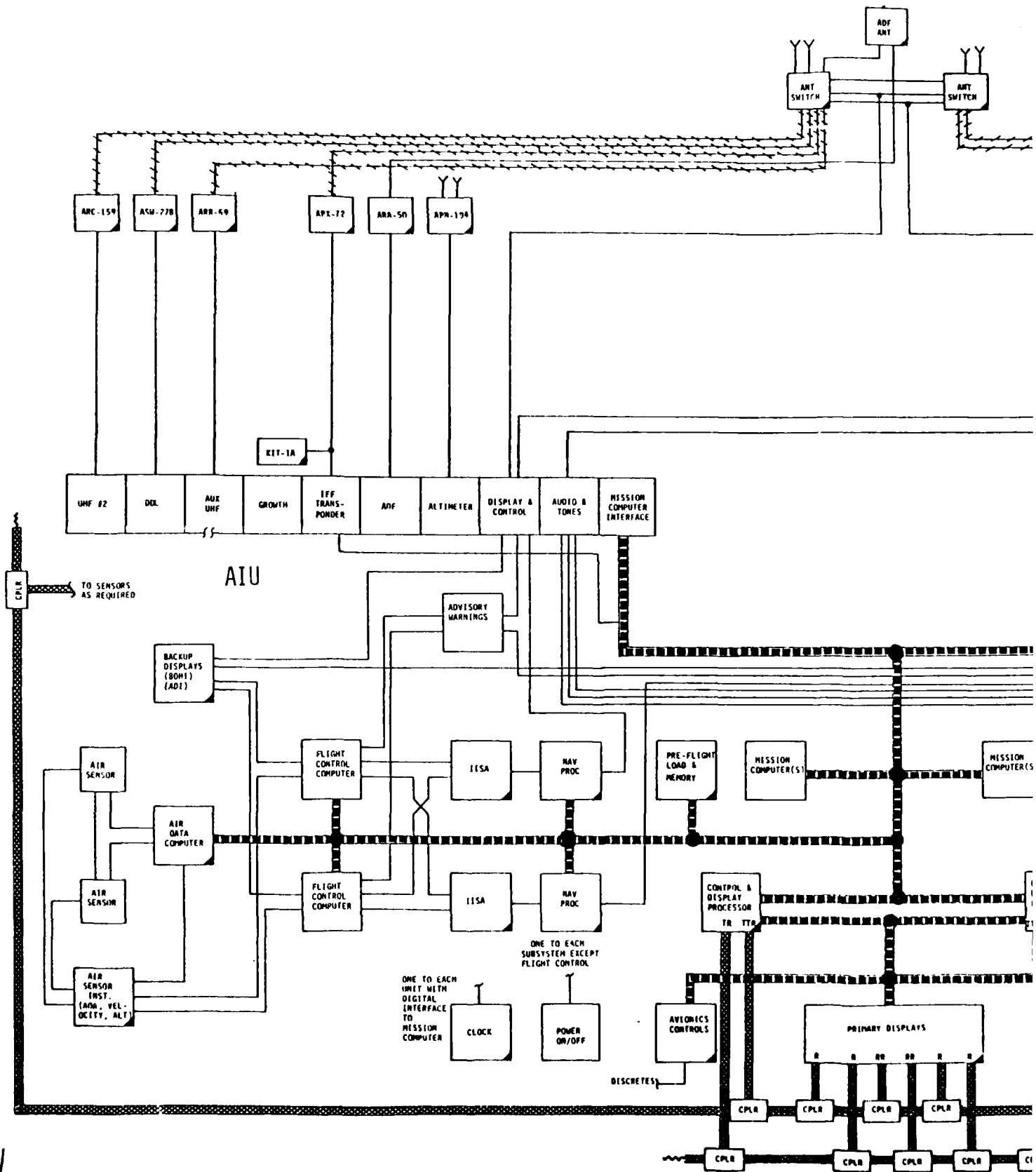
6.4.2.4.3 Configuration

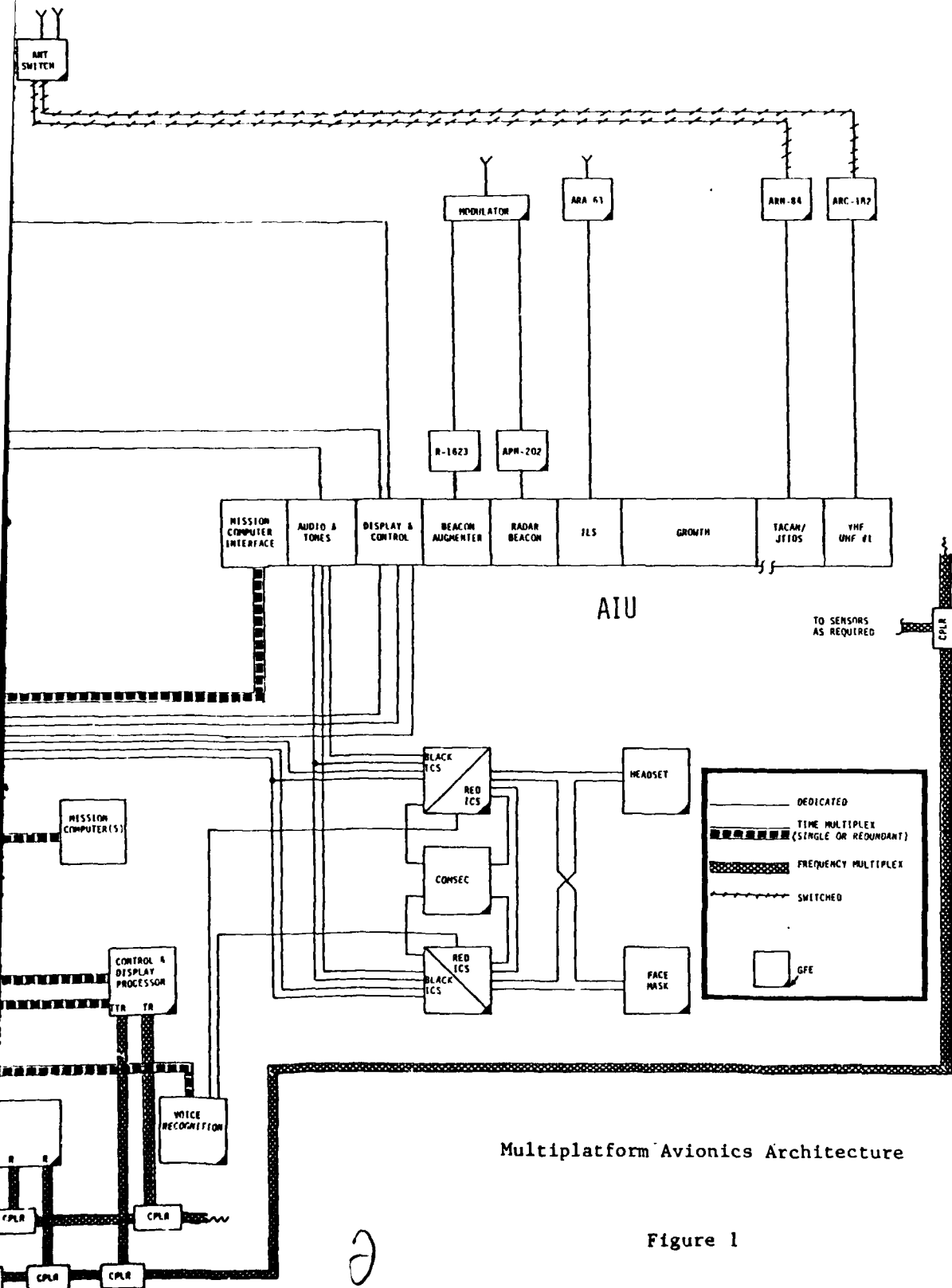
The clock unit is assumed to be an integral part of the Display and Control Subsystem with time of day display provided by that subsystem. The clock element could be part of the Display and Control Processor, with only one processor actually utilized to derive external sync outputs.

6.4.3 Technology Development

6.4.3.1 IF Multiplex Switch

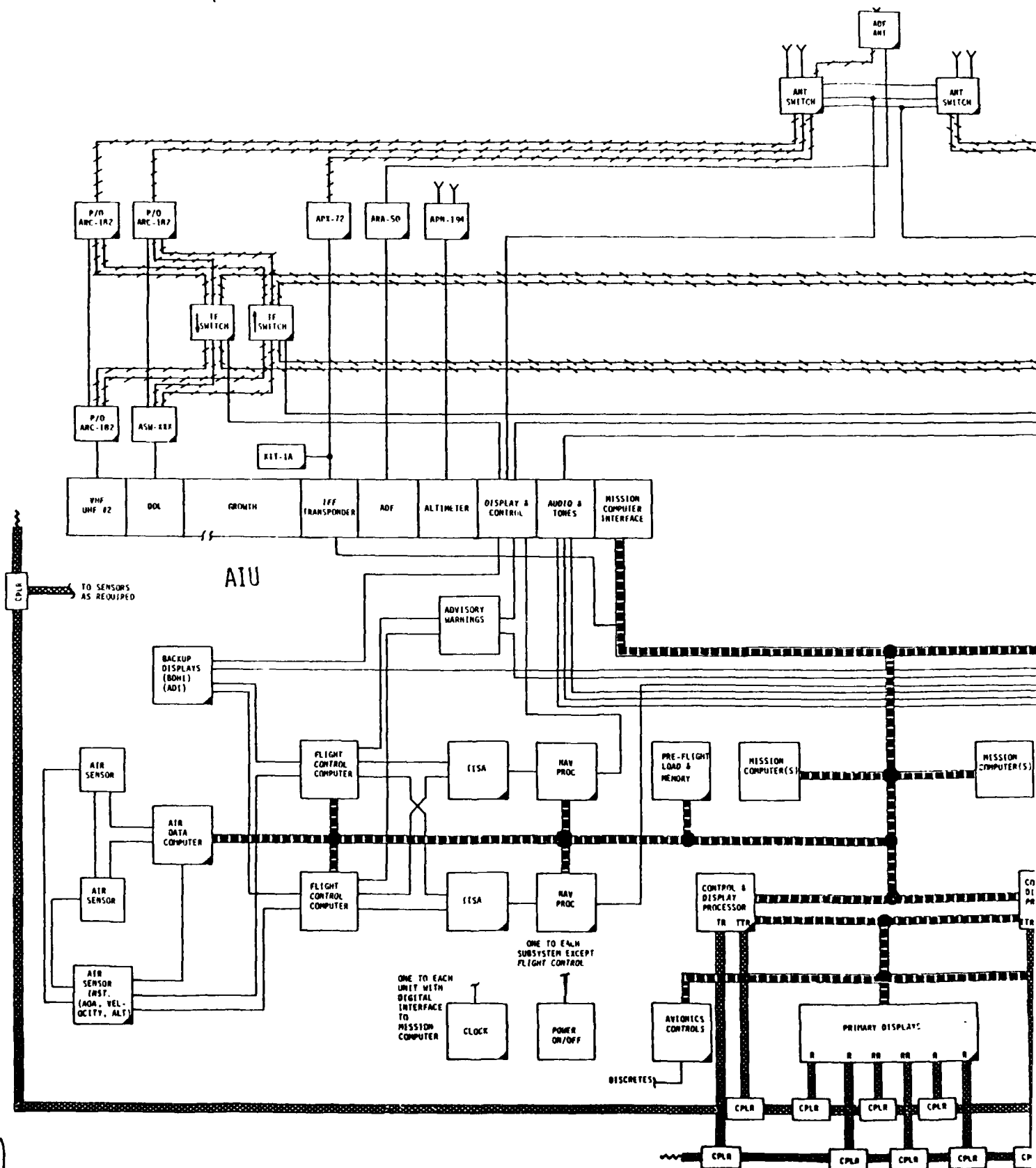
As discussed in 4.3.4 and 5.2 the development of a 70 MHz IF multiplex switch is recommended. The recommended configuration is a four by four matrix to provide unidirectional signal distribution. It is recommended that this be a self contained unit with fail safe input to output connections and digital control via the digital interface recommended in 6.2.1.2.

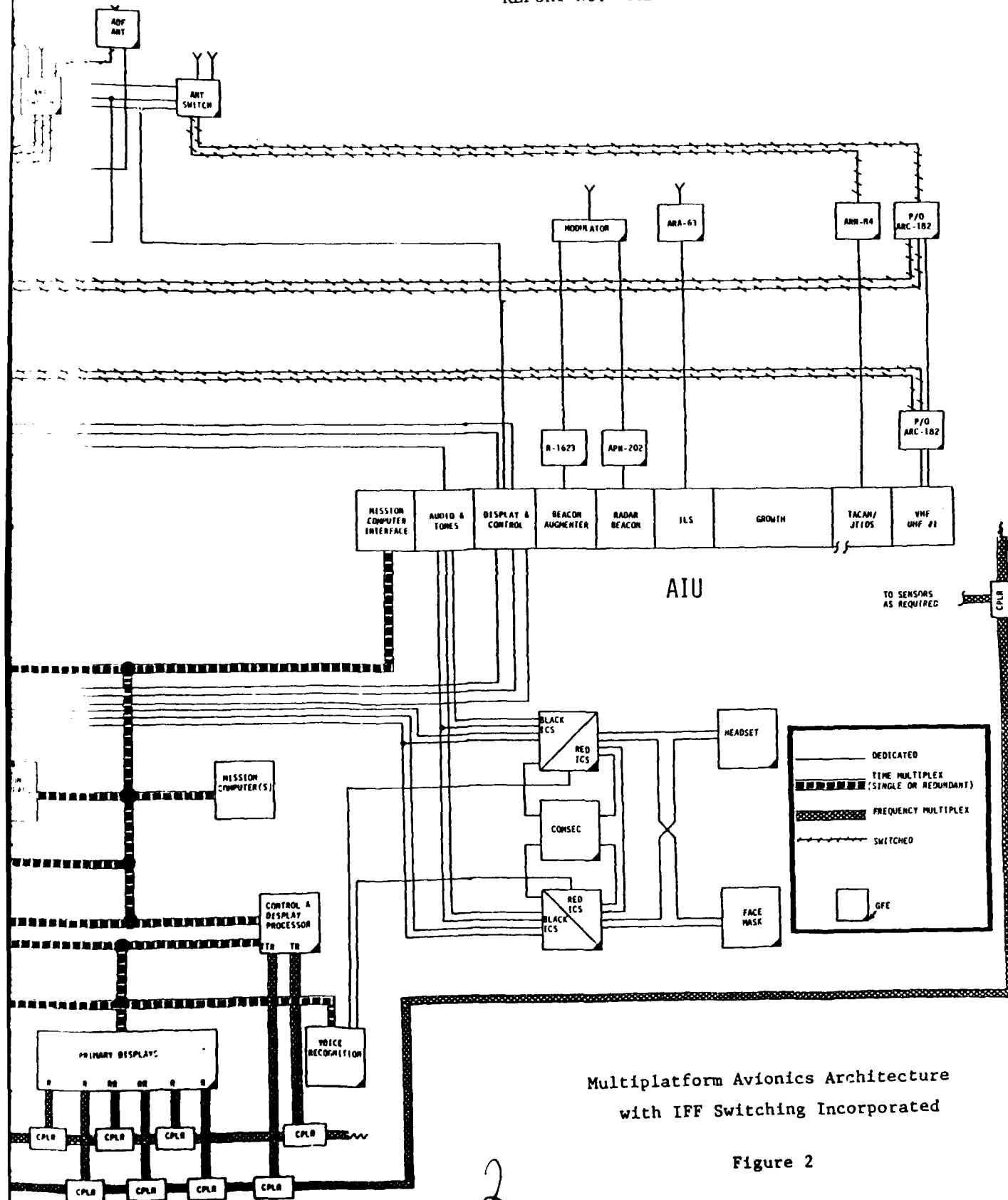




Multiplatform Avionics Architecture

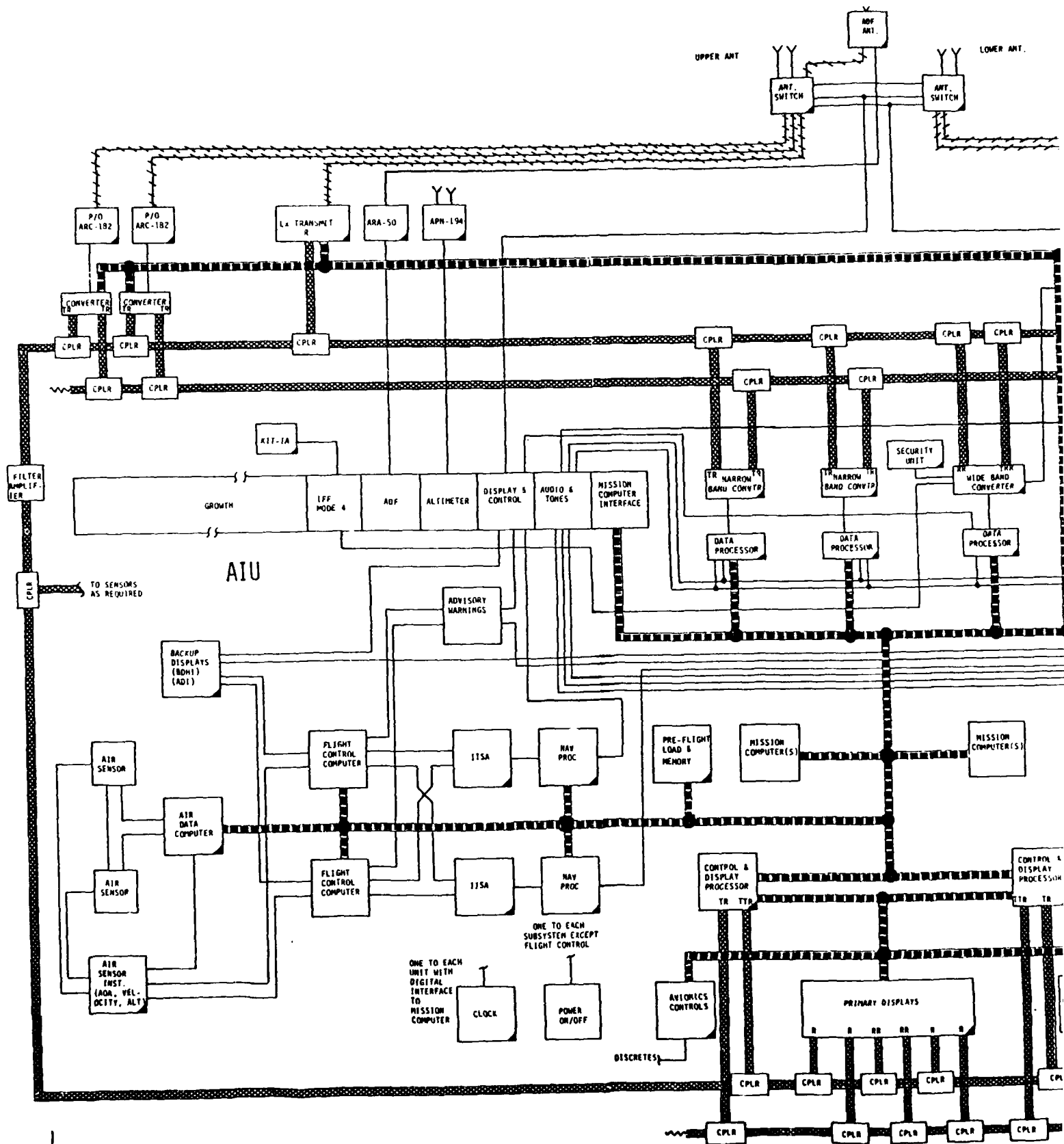
Figure 1

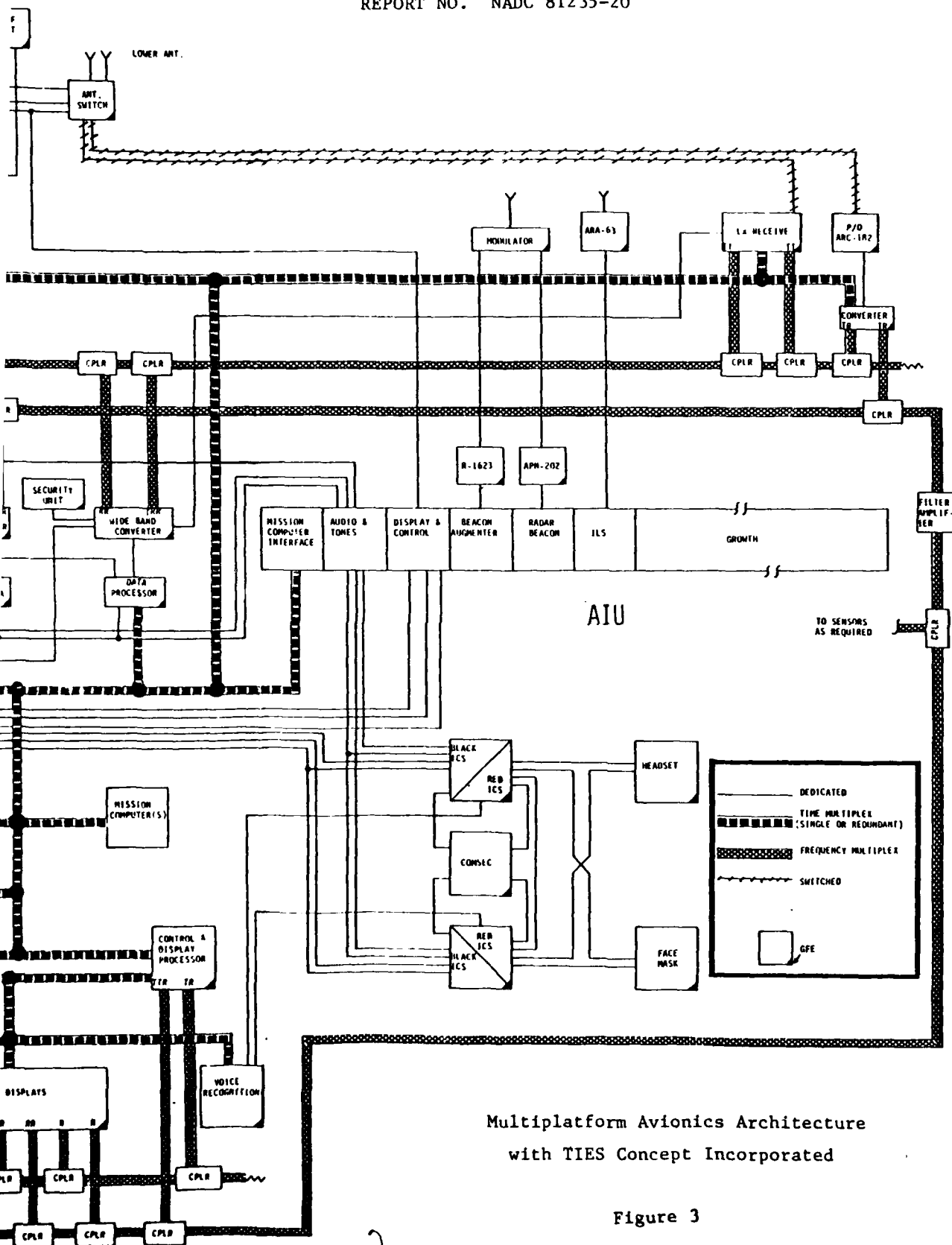




Multipatform Avionics Architecture
with IFF Switching Incorporated

Figure 2





Multiplatform Avionics Architecture
with TIES Concept Incorporated

Figure 3

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4 May 1981

APPENDIX A
SUBSYSTEM ARCHITECTURE, UNIT
CONFIGURATION AND THE
RELIABILITY IMPACT

Subsystem Architecture, Unit Configuration and the Reliability Impact

Introduction

The purpose of this analysis is to determine the effect on reliability of adding series elements to a subsystem architecture. The intent is to generate a means of assessing the reliability impact of adding additional units to the subsystem configuration, particularly for flight safety critical subsystems where parallel redundancy is incorporated. The configurations considered are shown in figure 1. The series configuration will be considered only as a reference point from which to evaluate series parallel flight safety critical subsystems. In this analysis all elements of the architecture will be considered active (i.e. operating, even though some or all outputs are not utilized) over the entire operating time of the subsystem and element interconnects will be considered failure free. Additionally, for simplicity and in order to allow generalized conclusions all elements are assumed to have identical failure (hazard) rates, and an exponential density function i.e.:

$$p(t) = e^{-\lambda t} \quad , \quad (1)$$

where $p(t)$ is the probability of success as a function of time and λ is the constant failure rate (failures per unit time) which is equal to $1/\theta_e$ where θ_e is the Mean Life of the element.

Care should be taken in interpreting the data that follows in that no attempt is made to account for the fact that after a repair a subsystem is assumed to be in the same status as a new system (i.e. no time deterioration). This assumption is consistent with the exponential density function in that all failures are chance failures and have a constant hazard rate (instantaneous failure rate) over all time. This assumption is not true for general avionics equipment (e.g. cooling fans wear out). The implied equivalence between MTBF (Mean Life Between Failures) of actual equipment and the calculated Mean Life in what follows is not accurate since wear out is not accounted for.

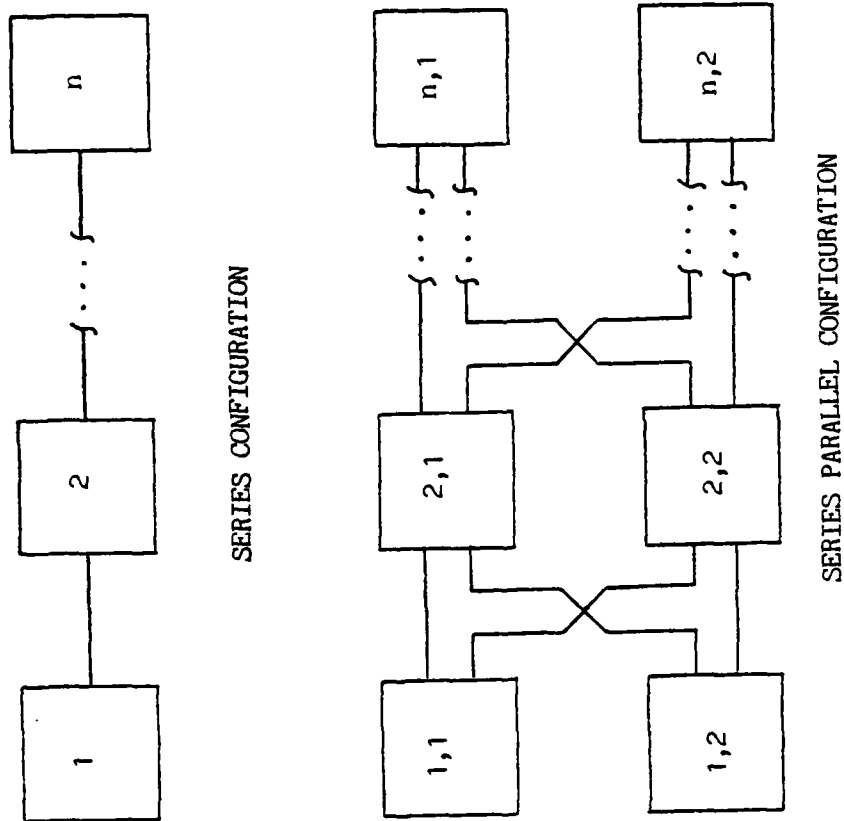


Figure 1. Architectural configurations.

Series Configuration

The Reliability Function for a subsystem with a series configuration of identical failure rate elements is:

$$R_c(t) = p(t)^n = e^{-n\lambda t} \quad (2)$$

The Mean Life is defined as:

$$\theta_c = \int_0^{\infty} R_c(t) dt = \int_0^{\infty} e^{-n\lambda t} dt = \frac{1}{n\lambda} \quad (3)$$

The effect on the Mean Life of a cascaded element subsystem due to the addition of one additional element may be expressed as:

$$\frac{\theta_c(n+1)}{\theta_c(n)} = \frac{n\lambda}{(n+1)\lambda} = \frac{n}{n+1} \quad (4)$$

Series Parallel Configuration

From table 8.2 item 4.c of the referenced literature a subsystem with a series parallel configuration of two identical parallel active elements with an exponential failure density has a Reliability Function given by:

$$R_s(t) = (p(t)^2 - 2p(t))^n \quad (5)$$

The Mean Life is:

$$\theta_s = \int_0^{\infty} R_s(t) dt \quad (6)$$

Substituting (1) in (5) and (5) in (6) and performing the integration yields:

$$\theta_s = \frac{1}{\lambda} \sum_{j=0}^n \frac{(-1)^j (2)^{n-j} n!}{(n+j) j! (n-j)!} \quad (7)$$

The values of $\lambda\theta$ for both the series and series parallel configurations equations of figure 1 (equations (3) and (7)) are plotted as continuous functions of n for illustrative purposes in figure 2. It should be noted that n may have integer values only and these functions are not continuous. Figure 2 illustrates the improvement in Mean Life of the series parallel configuration relative to the series configuration and the reduction in Mean Life due to the addition of elements to a subsystem configuration. In interpreting figure 2 as well as equations (5) and (7) it should be noted that Mean Life is defined in relation to a subsystem failure. For the series configuration a subsystem failure is synonymous with an element failure, however, for the series parallel configuration equations (5) and (7) and hence figure 2 assume no repair is made until a sufficient number of elements fail to cause a subsystem failure.

Series Parallel Configuration With Post Flight Maintenance

In section 8.6 of the referenced literature for an active series parallel subsystem for which maintenance is performed at intervals of time T the Reliability Function is given as:

$$R_m(t) = [R_s(T)]^j (R_s(\tau)), \text{ where} \quad (8)$$

$$t = jT + \tau$$

$$j = 0, 1, 2, \dots$$

$$0 \leq \tau < T$$

and $R_s(\gamma)$ is the Reliability Function of the subsystem without maintenance (see equation 5). The Mean Life of the subsystem is given by:

$$\theta_m = \int_0^{\infty} R_m(t) dt = \left[\sum_{j=0}^{\infty} R_s(T)^j \right] \int_0^T R_s(\tau) d\tau \quad (9)$$

and by using the identity $\sum_{j=0}^{\infty} x^j = \frac{1}{1-x}$ the Mean Life is restated as:

$$\theta_m = \frac{\int_0^T R_s(\tau) d\tau}{1 - R_s(T)} \quad (10)$$

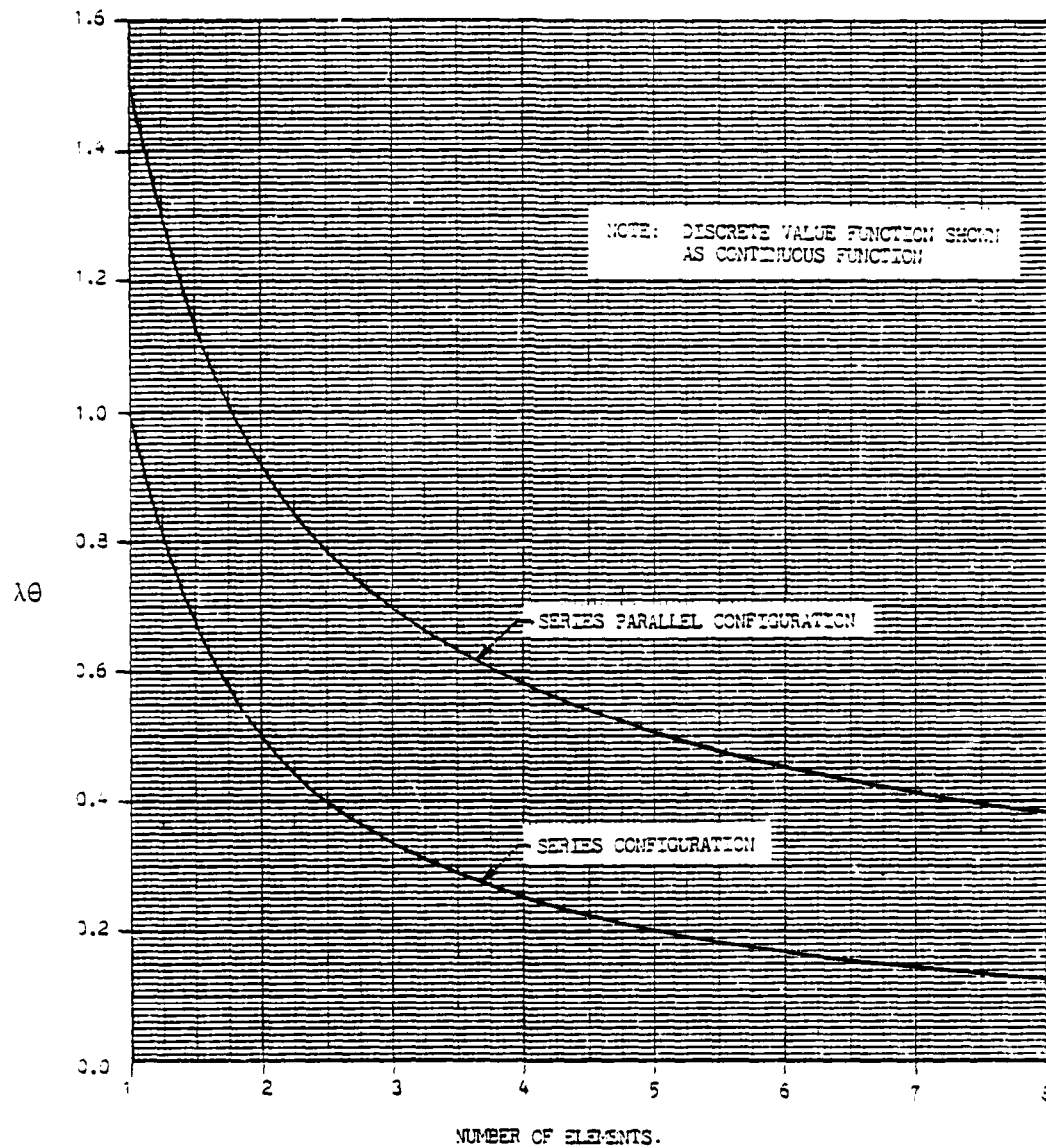


Figure 2. Ratio of the Mean Life of a configuration to the Mean Life of the elements versus the number of elements.

Substituting (1) in (5) and (5) in (10) yields:

$$\theta_m = \frac{\int_0^T (2e^{-\lambda\tau} - e^{-2\lambda\tau}) d\tau}{1 - (2e^{-\lambda T} - e^{-2\lambda T})} \quad (11)$$

Equation (11) has been evaluated as:

$$\theta_m = \frac{\frac{1}{\lambda} \sum_{j=0}^n \frac{(-1)^j (2)^{n-j} n!}{(n+j)! j! (n-j)!} (1 - e^{-(n+j)\lambda T})}{[1 - e^{-n\lambda T} (2 - e^{-\lambda T})^n]} \quad (12)$$

By inspection, as T approaches ∞ (i.e. no maintenance before a subsystem failure) equation (12) becomes identical to (7):

Figure 3 is a plot of the ratio:

$$\frac{\lambda \theta_m(n)}{\lambda \theta_m(n=1)} = \frac{\theta_m(n)}{\theta_m(n=1)}$$

as a function of n for three values of λT .

Again, for illustrative purposes figure 3 is plotted as a continuous function despite the fact that n is restricted to integer values. As noted above as T approaches ∞ (equivalent to $\lambda T \gg 1$) θ_m is equivalent to θ_s for a two element series parallel configuration of elements, hence the trace in figure 3 is equivalent to the two element series parallel trace of figure 2 with the maximum value normalized to 1. It is interesting to note that for $\lambda T \ll 1$ (i.e. maintenance in intervals much less than the MTBF of the subsystem elements) the sensitivity of the series parallel configuration to the number of elements (lower trace of figure 3) is identical to that of the non-redundant series configuration (lower trace of figure 2). It should be noted however that only the sensitivity ratios are identical, the absolute values vary drastically.

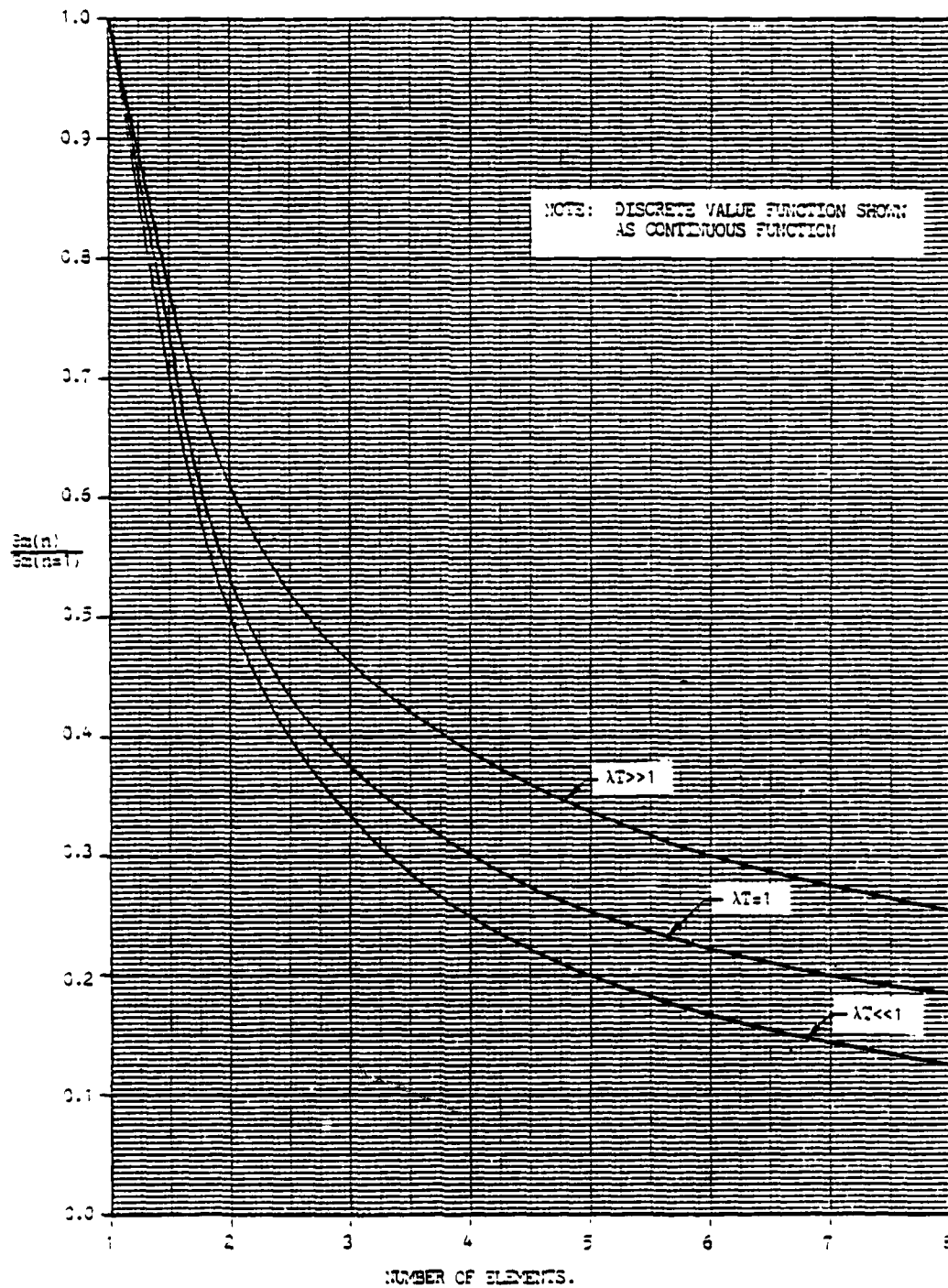


Figure 3. Effect on Mean Life of a series parallel configuration due to the addition of series element pairs.

(For example, a single element subsystem (series) with an element Mean Life of 100 hours has a subsystem Mean Life of 100 hours; a two element parallel subsystem with an element Mean Life of 100 hours has a subsystem Mean Life of 150 hours if no maintenance is performed until a subsystem failure occurs and a Mean Life of 5100 hours if after each 2 hour flight a redundant element failure is corrected.)

Discussion

Examination of figure 3 illustrates that when maintenance is considered the proportional effect on subsystem reliability is considerably greater than would be predicted by typical redundant system calculations. Figure 4 is a plot of the ratio $\frac{\Theta_m(n+1)}{\Theta_m(n)}$ using equation (12) to generate the values of $\Theta_m(n)$ and $\Theta_m(n+1)$. The figure shows the incremental effect of adding one more element to a subsystem of n elements. The lower trace ($\lambda T \ll 1$) is characteristic of a real world avionics system where the MTBF is significantly greater than a typical flight duration (i.e. maintenance period) and again interestingly is identical to a plot for the series configuration (equation (4)). Prior to attempting this analysis an arbitrary criteria that not more than a ten percent reduction in the MTBF of a subsystem should be tolerated to accommodate the addition of a single series element (with parallel redundancy) was established. This criteria can be met only for a series parallel subsystem configuration that consists of 9 or more element pairs when the effects of maintenance are considered. Since the sensitivity to adding one series element with parallel redundancy asymptotically approaches 1 any criteria more stringent than 10% would not allow the addition of a single element pair unless the subsystem was comprised of a large number of element pairs.

Conclusion

The addition of elements to a series parallel redundant subsystem architecture should be considered as if there was no redundancy. Although the reliability achieved by the series parallel redundant configuration is significantly greater than the series configuration the sensitivity to the number of elements is the same when real world maintenance

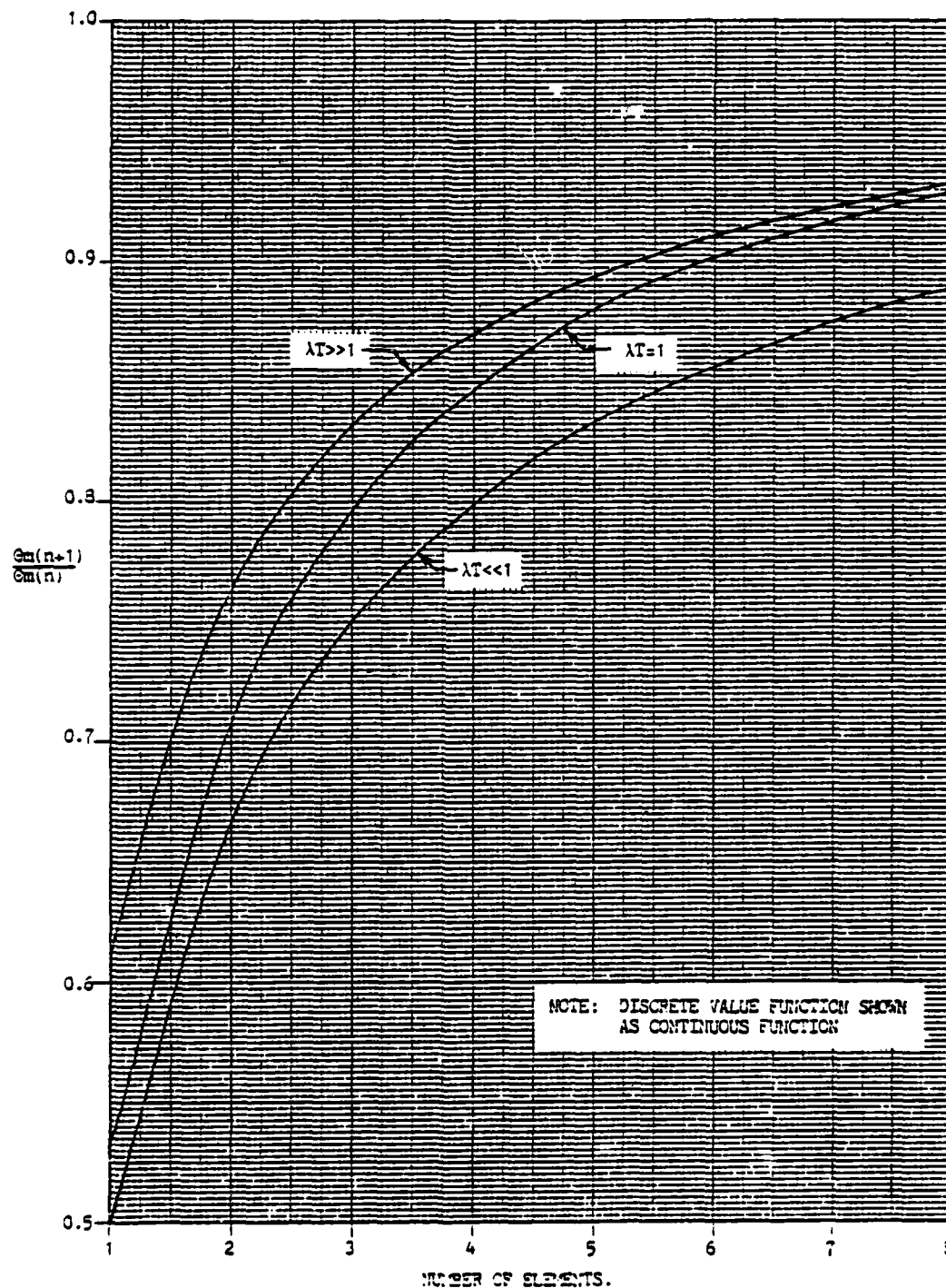


Figure 4. Incremental effect on Mean Life of adding elements to a series parallel configuration.

practices for avionics system are considered.

Figure 5 shows the ratio of, Mean Life for a series parallel subsystem with repair at intervals significantly less than the element Mean Life, to the Mean Life of the same subsystem with no repair until subsystem failure. The reliability increase achieved by repair of failed elements after each flight can only be realized if fault detection to indicate element failure is incorporated. Figure 5 emphasises that reliable fault detection and reporting should be given special attention in any architectural design.

References

Reliability Engineering, Arinc Research Corporation, edited by William H. Von Alven ©1964

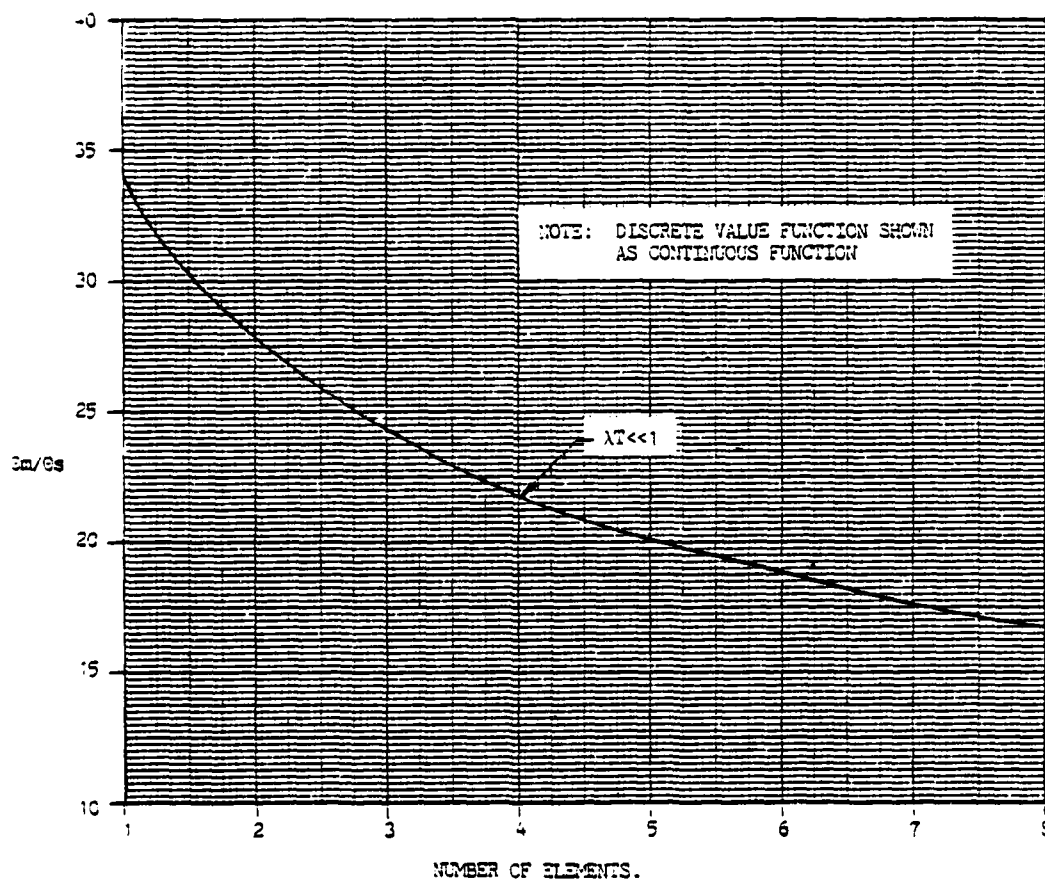


Figure 5. Improvement in Mean Life achieved by post flight maintenance of a series parallel configuration versus number of series elements.

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APPENDIX B

- o DEFINITION OF TERMS
- o GLOSSARY

DEFINITION OF TERMS

A number of terms utilized in this report have become diluted by application of more than one meaning. The definition of these terms as they are utilized in this report are as follows:

Weapons System. An entire military platform consisting of the vehicle, the crew, consumables, systems and subsystems required to perform one or more military missions.

System. That portion of the weapons system required to perform a specific task. Example: bombing system includes navigation, targeting, and weapons release/guidance.

System Avionics. That portion of a system primarily associated with collecting and processing information in electronic formats. Example: navigation processing and target tracking.

Weapons System Avionics. All the components of each of the Avionics Systems incorporated in a weapons system. The Weapons System Avionics must be considered as an entity in order to account for the requirements of shared equipment. Example, the Heads Up Display is part of the Bombing System as well as the Flight Control System.

Subsystem. A stand-alone entity that performs a specific function in support of one or more systems or avionics systems. Example: inertial navigation or radar.

DEFINITION OF TERMS (continued)

Specialized Subsystem. A subsystem that has peculiar requirements that do not allow complete integration into an avionics system. Example: manually updated doppler navigation or manually operated radar target tracking.

Avionics Architecture. The definition of the requirements necessary to perform all electronic information processing tasks in terms of equipment required, functional allocations for equipment, and how the equipment interface with each other and the operator. This definition may apply to weapons system avionics, system avionics, or an individual subsystem.

Multiplatform Avionics Architecture. That portion of the Weapons System Avionics Architecture designated to be common among platforms. It includes all elements of the Weapons System Avionics Architecture except the specific equipment (subsystems) to be utilized, functional allocations for individual elements of a subsystem, interfaces between elements of a specialized subsystem. Functional allocations for equipment (subsystems) are included only to the extent of defining what will not be performed by the subsystems.

Core Avionics. The avionics systems whose functions are universally applicable among most aircraft types. These systems include voice communications, data communications, navigation systems and aids, portions of the flight control system and avionics controls and displays.

DEFINITION OF TERMS (continued)

System Engineering. The design of a complex interrelation of many elements (a system or subsystem) to achieve a performance requirement, taking into consideration all the elements (including the operator/user) related in any way to the system or subsystem. The result of the design effort is a system or subsystem configuration and a functional allocation for each element of the system or subsystem.

Systems Integration. The process of interconnecting a subsystem to a weapons system in such a manner as to achieve greater capability than is inherent in either the original system or any of the interconnected subsystems.

GLOSSARY

AAES	Advanced Avionics Electrical System
ABC	fictitious contractor
ac	alternating current
ACLS	Automatic Carrier Landing System
ADF	Automatic Direction Finding
ADI	Attitude Director Indicator
AIDS	Advanced Integrated Display System
AIU	Avionics Interface Unit
AM	Amplitude Modulation
AOA	Angle of Attack
ARI	Attitude Reference Indicator
ARINC	Aeronautical Radio Incorporated
AVCS	Avionics Components and Subsystems
A/D	Analog to Digital
BC	Bus Controller
BDHI	Bearing Distance Heading Indicator
BIT	Built in Test
CAINS	Carrier Aircraft Inertial Navigation System
CFE	Contractor Furnished Equipment
COMSEC	Communications Security
CPLR	Coupler
CSC	Communications System Control
CSDC	Computer Signal Data Converter
CV	Aircraft Carrier
CVS	Correlation Velocity Sensor
dB	deci Bell
dBm	deci Bell relative to a milliwatt
dc	direct current

GLOSSARY (continued)

DDL	Digital Data Link
DECM	Deceptive Electronic Countermeasures
DOD	Department of Defence
DSB	Double Side Band
EIA	Electronic Industries Association
EMI	Electromagnetic Interference
ESM	Electronic Support Measures
EW	Electronic Warfare
FED	Federal
FLIR	Forward Looking Infrared Receiver
FM	Frequency Modulation
FSK	Frequency Shift Keying
GFE	Government Furnished Equipment
GFS	Government Furnished Software
GHz	giga Hertz
GPS	Global Positioning System
GSE	Ground Support Equipment
HF	High Frequency
HSI	Horizontal Situation Indicator
Hz	Hertz
ICS	Intercommunications System
IEEE	Institute of Electrical and Electronic Engineers
IF	Intermediate Frequency
IFF	Identification Friend or Foe
IFPM	In-flight Performance Monitoring
IISA	Integrated Inertial Sensor Assembly

GLOSSARY (continued)

ILS	Instrument Landing System
I/O	Input/Output
JTIDS	Joint Tactical Information Distribution System
KHz	kilo Hertz
LCC	Life Cycle Cost
LF	Low Frequency
LF/ADF	Low Frequency/Automatic Direction Finding
LORAN	Long Range Aid to Navigation
Lx	960 to 1215 MHz band
MESM	Mission Essential Subsystem Matrices
MHz	mega Hertz
MIL	Military
MLS	Microwave Landing System
MMR	Multimode Receiver
msec	millisecond
NAC	Naval Avionics Center
NAVAIR	Naval Air Systems Command
NAVAIRDEVCEN	Naval Air Development Center
NRZ	Non Return to Zero
nsec	nanosecond
NTDS	Navy Tactical Data System
OMEGA	not an acronym
OPNAV	Offices of the Chief of Naval Operations
OPNAVINST	OPNAV Instruction
PCFE	Platform Contractor Furnished Equipment
PM	Phase Modulation

GLOSSARY (continued)

PROTEUS	not an acronym
PRP	Pulse Repetition Period
QPL	Qualified Parts List
R	Receive
RF	Radio Frequency
RT	Remote Terminal
R/T	Receive/Transmit
SAE	Society of Automotive Engineers
SEMP	Standard Electronic Module Program
STD	Standard
T	Transmit
TACAN	Tactical Air Navigation
TEMPEST	not an acronym
TIES	Tactical Information Exchange System
UHF	Ultra High Frequency
V	Volt
VHF	Very High Frequency
VHSIC	Very High Speed Integrated Circuits
VLF	Very Low Frequency
VOR	VHF Omni Range
VOR/LOC	VOR/Localizer
XYZ	fictitious contractor
usec	microsecond

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