

AD-A117 136

ROYAL AIRCRAFT ESTABLISHMENT FARNBOROUGH (ENGLAND)

F/6 11/4

A NEW TETRA-EPOXIDE RESIN AS A MATRIX FOR ADVANCED COMPOSITES. (U)

NOV 81 L N PHILLIPS, D J MURPHY

UNCLASSIFIED

RAE-TM-MAT-380

DRIC-BR-82008

NL

1 of 1  
AD-A  
17136



END  
DATE  
FILMED  
08-82  
DTIC

TECH. MEMO  
MAT 380

BR82008

TECH. MEMO  
MAT-380



AD A117136

UNLIMITED

ROYAL AIRCRAFT ESTABLISHMENT



A NEW TETRA-EPOXIDE RESIN AS A MATRIX FOR ADVANCED COMPOSITES

by

L. N. Phillips

D. J. Murphy

November 1981

DTIC FILE COPY

DTIC  
ELECTE  
JUL 21 1982  
S D  
E

82 07 19 186

UNLIMITED

ROYAL AIRCRAFT ESTABLISHMENT

Technical Memorandum Mat 380

Received for printing 23 November 1981

A NEW TETRA-EPOXIDE RESIN AS A MATRIX FOR ADVANCED COMPOSITES

by

L. N. Phillips

D. J. Murphy

SUMMARY

Two versions of a new tetra-epoxide resin were supplied by Shell Laboratories, for evaluation as a possible matrix for carbon fibre.

Because of the absence of tack in the resulting pre-preg both resins were modified at RAE with a minor percentage of the Bisphenol A resin, Shell 828.

All four formulations (*ie* modified and unmodified 114 and 115 resins) had their curing characteristics established and were then used in conjunction with Courtaulds XAS carbon fibre, the reinforcement being in the form of satin-weave and unidirectional fabrics.

The mechanical properties of the derived laminates were obtained at room temperature, at 80°C and 130°C. Resistance to a range of aircraft fluids was also examined.

The best results overall were obtained with the lower molecular weight epoxide blended with Bisphenol A resin; and it is recommended that more extensive trials and long-term testing should begin.

*Copyright*

©

*Controller HMSO London*

*1981*

LIST OF CONTENTS

	<u>Page</u>
1 INTRODUCTION	3
2 MATERIALS USED	3
2.1 Resins and hardeners	3
2.2 Carbon fibre reinforcements	4
3 MANUFACTURE OF PRE-PREG	4
4 USE OF TORSIONAL BRAID ANALYSIS (TBA) TO FOLLOW CURE OF RESINS	4
5 PREPARATION OF LAMINATES	5
6 TESTING PROCEDURES	5
7 RESULTS OF MECHANICAL TESTS	6
8 RESISTANCE TO AIRCRAFT FLUIDS	7
9 DISCUSSION AND CONCLUSIONS	7
Acknowledgments	9
References	9
Tables 1 to 5	10
Illustrations	Figures 1&2
Report documentation page	inside back cover

<b>Accession For</b>	
NTIS GRA&I	<input checked="" type="checkbox"/>
DTIC TAB	<input type="checkbox"/>
Unannounced	<input type="checkbox"/>
Justification	
By _____	
Distribution/	
Availability Codes	
Dist	Avail and/or Special
<b>A</b>	



## 1 INTRODUCTION

As part of the RAE programme on the development and evaluation of potential matrix resins for composite materials, an extra-mural contract was placed with the Shell Chemical Company in 1979.

The objective was to seek inexpensive alternatives to the epoxide compound tetra-glycidyl-4,4'-diaminodiphenyl methane (which is the basis for many of the resin systems presently used by the aircraft industry<sup>1</sup>) without substantial sacrifice either of ease of processing, or of mechanical properties, of the resulting laminates.

Also of importance were the resistance to the range of normal aircraft fluids and the amount of water uptake at saturation, which has an effect on compressive performance under hot, wet conditions<sup>2</sup>.

After preliminary synthetic work at Shell, a new tetra-epoxide was selected as a suitable candidate; and two versions, in a higher and lower molecular weight were submitted to Materials Department for examination. The resins are designated as Shell 1153/114 for the low molecular weight and Shell 1153/115 for the higher molecular weight materials\*. Both are pale yellow, brittle solids as received, freely soluble in acetone, methyl ethyl ketone, etc.

Stoichiometric amounts of diaminodiphenyl sulphone, which is well-known as a heat-resistant aromatic amine, were used as the hardener, with a small addition of boron trifluoride etherate as catalyst.

The reinforcement of particular interest is carbon fibre. It was used in the form of standard unidirectional and satin-weave fabrics, based on Courtaulds XAS grade for which data are available with other resin systems<sup>3</sup>.

In the reconnaissance reported here, the curing and moulding behaviour of the Shell resins, both unmodified and modified, is reported, together with the mechanical properties of the derived laminates.

## 2 MATERIALS USED

### 2.1 Resins and hardeners

The two experimental tetra-epoxide resins, designated 114 and 115, were supplied by Shell Chemicals as solid castings which could be easily broken up into small pieces convenient for dissolving in acetone. The lower molecular weight resin 114, has an epoxide equivalent weight of 212, whilst that of the 115 system is 235. Melting points of the two resins are 38-43°C for the 114 resin and 64-69°C for the 115 material.

In addition to the two principal resins used, the well known DGEBA resin Epikote 828 supplied by Shell Chemicals, was used as a modifier.

The curing agent used throughout the programme was diaminodiphenyl sulphone (DDS) supplied by Koch-Light Laboratories Limited. Shell Chemicals Epikure BF<sub>3</sub>400, boron trifluoride monoethylamine complex, was used as a catalyst.

---

\* These will be called 114 and 115 resins throughout the text.

## 2.2 Carbon fibre reinforcements

Two carbon fibre fabrics were chosen, a five-shaft satin and a tied unidirectional type. Both cloths were based on Courtauld's 3000 filament XAS carbon fibre and were woven by Carr Reinforcements Limited. The unidirectional material was held together with an 11 tex glass fibre weft; this being preferred as the result of a previous investigation<sup>4</sup>.

## 3 MANUFACTURE OF PRE-PREG

Solutions of the resins were prepared by dissolving the appropriate amount of resin and hardener in acetone so as to give a 40% w/w solution. The amount of DDS curing agent required was slightly different for each resin, this being dependent on the epoxide content.

The carbon fibre fabrics were impregnated with the solutions by brush coating. After being allowed to stand for 3 h the pre-preg was placed in an air-circulating oven, set at 100°C, for 20 min. The pre-pregs produced had a resin solids content of 40% by weight.

The 114 and 115 resins, being solids at room temperature, were found to produce pre-pregs with little or no tack.

In an attempt to improve the tack and flexibility of the pre-pregs produced, increasing proportions of the liquid DGEBA resin Epikote 828 were added to the prime resin. It was found that a 33% by weight addition was needed to give a satisfactory pre-preg.

The detailed formulations used for the modified and straight resins are given below:

114 resin	114-828 blend	115 resin	115-828 blend
100 g 114	100 g 114	100 g 115	100 g 115
30 g DDS	50 g 828	27 g DDS	50 g 828
1 g BF <sub>3</sub> 400	47 g DDS	1 g BF <sub>3</sub> 400	42 g DDS
	1.5 g BF <sub>3</sub> 400		1.5 g BF <sub>3</sub> 400

## 4 USE OF TORSIONAL BRAID ANALYSIS (TBA) TO FOLLOW CURE OF RESINS

In order to establish the optimum cure conditions for the resins under evaluation, the Polymer Chemistry Section, Materials Department, RAE, carried out a series of isothermal runs to determine the gel and vitrification times at various temperatures. The TBA results are presented in Figs 1 and 2. It is clear from these results that the higher molecular weight resin, 115 cures more rapidly than the 114 system and that the addition of the Epikote 828 resin slows down the cure of both the 114 and 115 systems. The TBA work also showed that if the modified resins were post-cured at a temperature of 190°C, they both achieved a glass transition temperature (T<sub>g</sub>) of between 241 and 245°C after 12 h. It was found however that the T<sub>g</sub> had reached about 95% of this value after only 2 h post cure.

## 5 PREPARATION OF LAMINATES

All the laminates were compression moulded in a light-alloy tool of internal dimensions 315 mm × 215 mm. The moulding tool was coated with Vydax release agent prior to use.

The moulding procedure was the same for both the satin-weave and unidirectional fabric and is outlined below.

The pre-preg materials were cut to size and the requisite number of layers placed in the cold mould; this being eight for the satin-weave and 15 for the unidirectional fabric. Each layer of satin-weave pre-preg was stacked in the mould with the warp fibres along the 0° axis.

The loaded mould was placed in an electrically heated hydraulic press, the temperature of the platens being maintained at 150°C. The moulding was allowed to dwell for some minutes with only contact pressure being applied; this time varied between 12 min for the 115 system, and 18 min in the case of the 114-828 formulation. After the dwell period the mould was pressed down to 1.9 mm stops in the case of the satin-weave fabric and 2.0 mm for the unidirectional fabric, a pressure of approximately 2 MPa being required for this.

The mould was left in the press for a further 1 h before turning the heaters off and allowing it to cool. Once the temperature had dropped below 60°C the mould was removed from the press, and the laminate ejected from it.

All the laminates were subjected to a post-cure. The schedule followed was, 2 h at 170°C plus 16 h at 190°C in the case of the 114 based laminates, and 16 h at 190°C for 115 based laminates.

## 6 TESTING PROCEDURES

Determinations of flexural strength, tensile strength, tensile modulus and interlaminar shear strength were all performed at room temperature, and in addition, the flexural and interlaminar shear strengths were determined at 80°C and 130°C. The test specimens were cut from moulded sheets using a diamond-edged cutting wheel (152 mm diameter × 1.6 mm thickness), the grit size being 85-100. The material was tested in the longitudinal (warp) direction on a floor-mounted Instron machine (type 1115). Ten replicate specimens were normally taken for each test. The details of the various mechanical tests performed are given below.

### (i) Flexural strength

A standard specimen, 100 mm × 10 mm, as described by Sturgeon<sup>5</sup> was used.

The specimen was loaded at a rate of 20 mm/min.

### (ii) Tensile modulus

A parallel sided specimen 200 mm × 10 mm was used. Light-alloy end-pieces were bonded to the ends of each specimen using Redux 410 epoxy adhesive.

The specimens were loaded at a rate of 2 mm/min to about 50% of their expected failing load and the strain measured using an Instron strain gauge extensometer.

(iii) Tensile strength

Once the tensile modulus of the satin-weave specimens had been determined they were tested to failure. The specimens made from the unidirectional fabric had a radius ground on them in accordance with the recommendations of Dootson<sup>6</sup>, before testing to destruction.

The load was applied at a rate of 2 mm/min.

(iv) Interlaminar shear strength

A standard specimen 12 mm × 10 mm, as described by Dootson<sup>6</sup> was used.

The rate of loading was 2 mm/min.

7 RESULTS OF MECHANICAL TESTS

The detailed room temperature properties are given in Tables 1 and 2, whilst Tables 3 and 4 show the effect of increasing temperature on the flexural and interlaminar shear strengths.

Considering the room temperature results first, both the 114 and 115 systems performed quite well. It is clear however, that the 114 systems are somewhat superior to those based on 115. The addition of the Epikote 828 resin to both 114 and 115, showed no adverse effect; indeed an *increase* in properties took place in most cases. This general increase in the mechanical properties with the addition of Epikote 828 may be associated with better adhesion of the matrix to the carbon fibre; the interlaminar shear values being higher in every case.

At 80°C the flexural strength of all the resin/fabric combinations held up well; 90% or more retention in all but one case. It is evident from both the flexural and interlaminar shear strengths at 80°C that the addition of Epikote 828 has given improved retention of properties. Little difference between the 114 and 115 resins was noted with regard to the percentage retention of properties.

When the temperature was increased to 130°C the flexural strength of the systems containing the 828 addition again showed improved retention of strength as compared with the base systems. The 114-828 resin reinforced with satin-weave fabric retained an excellent 93% of the room temperature value, although the same resin reinforced with the unidirectional material, retained only 73%. A similar difference between the satin-weave and the unidirectional fabric was found with the 114 system. It is interesting to note that in a previous evaluation of another resin system<sup>7</sup>, the unidirectional fabric was found to perform somewhat worse at elevated temperatures than, in that case, an unwoven unidirectional carbon fibre. It would appear that the presence of a small amount of glass weft may adversely affect the strength retention of CFRP at elevated temperatures.

The interlaminar shear values followed the same trends as those observed with flexural strength except that the reduction in properties was generally rather more marked. However, despite the reduction of values, the results are still quite acceptable even at 130°C; 56 MPa for example in the case of unidirectional fabric with the 114-828 matrix.

## 8 RESISTANCE TO AIRCRAFT FLUIDS

A small exercise was performed to see whether some of the commonly encountered aviation fluids were likely to have any serious effect on carbon fibre composite material, containing the 114-828 resin system.

Eight different fluids were selected, mostly chosen from British Standard 3G.100: Part 2: sub-section 3.12, and these are listed in Table 5. In addition to the eight aircraft fluids used, the effect of water was also assessed.

Small pieces of unidirectional composite 100 mm × 10 mm × 2 mm in size, were carefully weighed and then immersed in the various fluids which were kept at a temperature of 50°C. After 48 h the pieces of composite were removed from the fluid and the weight change recorded. The specimens were returned to the fluids and then removed and re-weighed after a total time of 7 days. Before the specimens could dry-out, their flexural strengths were also measured and these results are given in Table 5 together with the recorded weight changes.

These show that for most of the fluids tested, there was a negligible change in weight and an unchanged flexural strength as compared with the control. The greatest change was produced by water, which gave a weight pick-up of about 0.5% after 7 days immersion at 50°C accompanied by a 14.3% reduction in flexural strength.

## 9 DISCUSSION AND CONCLUSIONS

In assessing the value of a new resin system we are interested first in the absolute level of mechanical properties at room temperature as compared with the same reinforcement in standard matrices; and second in the reduction in mechanical properties with elevated temperature.

In the case of XAS fibre, information is available from previous work with the Ciba-Geigy XD 927 resin<sup>3,4</sup>, with the Friedel-Crafts system Xylok 237<sup>7</sup> and with the Shell pre-condensate system Epikote DX 210<sup>7,8</sup>. This information is summarised for unidirectional material at room temperature:

Matrix	Flexural strength (MPa)	Tensile strength (MPa)	Interlaminar shear strength (MPa)
114-828	2018	1480	92
XD 927	2027	1840	73
DX 210/BF <sub>3</sub> 400 (unwoven)	1919	1567	99

For satin-weave fabrics, laminates tested at room temperature gave:

Matrix	Flexural strength (MPa)	Tensile strength (MPa)	Interlaminar shear strength (MPa)
114-828	918	783	51
XD 927	1020	791	58

The extent of reduction in properties, as the temperature is raised from room temperature, through 80°C to 130°C is given by:

Matrix/fibre	Temperature	Flexural strength (% retention)
114-828/XAS	RT	100
	80°C	91
	130°C	73
DX 210/AS	RT	100
	80°C	88
	120°C	76

These results taken together show that the new resin system gives very similar room temperature figures to other resins with both kinds of woven reinforcement. At elevated temperatures, strength retention is rather better than for a Friedel-Crafts system<sup>7</sup>, comparing favourably with Epikote DX 210.

The following points may be made in conclusion:

- (1) Of the two versions of the tetra-epoxide supplied, the lower molecular-weight version, resin 114, is more convenient to use and gives better mechanical properties at room temperature and above.
- (2) The blend of 114 with 828 resin is better than the unmodified 114 alone, in respect of both tack and flexibility in the pre-preg form.
- (3) The 114-828 system is easy to handle, has a convenient cure and post-cure schedule and gives a final T<sub>g</sub> in the region of 240-245°C.
- (4) The strength and stiffness of laminates at room temperature is high, with good strength retention at elevated temperatures. Depending on the end use, an upper working limit of 130°C is possible.
- (5) There is good resistance to the normal range of aircraft fluids, with an acceptably low moisture absorption.
- (6) The 114-828 blend seems to be a very good general purpose pre-preg system for carbon fibre and it is recommended that full-scale trials be mounted both for pre-pregging on commercial equipment and for the necessary long-term testing in fatigue, heat-ageing, weathering and moisture resistance.

### Acknowledgments

The torsional braid analysis results were provided by Dr W.A. Lee and Mr M.J. Oliver, using the special automatic-recording TBA apparatus developed in the Polymer Chemistry Section, Materials Department.

### REFERENCES

- | <u>No.</u> | <u>Author</u>                     | <u>Title, etc</u>   |
|------------|-----------------------------------|---|
| 1          | Roger J. Morgan<br>Eleno T. Mones | Structure-property relations of composite matrices.<br>Proceedings of the ACS/CSJ Chemical Congress 'Resins for<br>Aerospace', pp 233-245, Hawaii 1979                      |
| 2          | J. Eastham                        | In-depth evaluation of unidirectional 1305C 10000/Fiberdux 914C<br>carbon fibre composites.<br>Final report on MOD Contract No.K/LR32B/2126                                 |
| 3          | L.N. Phillips<br>D.J. Murphy      | Evaluation of woven fabrics based on Grafil XAS, 3000-filament<br>carbon fibre.<br>RAE Technical Memorandum Mat 339 (1980)  |
| 4          | L.N. Phillips<br>D.J. Murphy      | The effect of twist and weft construction on the mechanical<br>properties of unidirectional fabric-based carbon fibre laminates.<br>RAE Technical Memorandum Mat 359 (1981) |
| 5          | J.B. Sturgeon                     | Specimens and test methods for carbon fibre reinforced plastics.<br>RAE Technical Report 71026 (1971)   |
| 6          | M. Dootson                        | Standard specimens and test methods for unidirectional carbon<br>fibre reinforced plastics.<br>British Aerospace Corporation, Warton,<br>Technical Report TN 4440 (1976)    |
| 7          | L.N. Phillips<br>D.J. Murphy      | Evaluation of carbon fibre laminates using an epoxide-hardened<br>Friedel-Crafts resin as the matrix.<br>RAE Technical Memorandum Mat 322 (1979)                            |
| 8          | L.N. Phillips<br>D. J. Murphy     | Grafil XAS laminates - an initial evaluation.<br>RAE Technical Memorandum Mat 298 (1978)  |

REPORTS QUOTED ARE NOT NECESSARILY  
APPROVED BY THE PUBLIC  
OR TO CONTROLLED ORGANISATIONS

Table 1

ROOM TEMPERATURE PROPERTIES OF LAMINATES MADE WITH THE VARIOUS  
RESIN SYSTEMS AND SATIN-WEAVE CLOTH

Resin	Flexural strength (MPa)	Tensile strength (MPa)	Tensile modulus (GPa)	Interlaminar shear strength (MPa)
114	Mean = 881 CV = 6.2%	Mean = 743 CV = 8.1%	Mean = 68 CV = 8.9%	Mean = 47 CV = 7.8%
114-828	Mean = 918 CV = 9.7%	Mean = 783 CV = 13.2%	Mean = 73 CV = 4.6%	Mean = 51 CV = 9.9%
115	Mean = 822 CV = 11.2%	Mean = 691 CV = 7.9%	Mean = 68 CV = 7.3%	Mean = 42 CV = 6.8%
115-828	Mean = 859 CV = 8.5%	Mean = 802 CV = 4.9%	Mean = 71 CV = 3.5%	Mean = 53 CV = 13.5%

$V_f = 0.60$ , CV = coefficient of variation

Table 2

ROOM TEMPERATURE PROPERTIES OF LAMINATES MADE WITH 114 AND 114-828  
RESIN SYSTEMS WITH UNIDIRECTIONAL CLOTH

Resin	Flexural strength (MPa)	Tensile strength (MPa)	Tensile modulus (GPa)	Interlaminar shear strength (MPa)
114	Mean = 2150 CV = 3.5%	Mean = 1631 CV = 7.8%	Mean = 131 CV = 2.6%	Mean = 88 CV = 11.1%
114-828	Mean = 2018 CV = 8.2%	Mean = 1480 CV = 5.5%	Mean = 131 CV = 4.3%	Mean = 92 CV = 5.2%

$V_f = 0.62$

Table 3

THE EFFECT OF INCREASING TEMPERATURE ON THE FLEXURAL STRENGTH OF LAMINATES  
MADE WITH 114 OR 115 BASED RESIN

Temperature of test °C	114 Satin-weave	114-828 Satin-weave	115 Satin-weave	115-828 Satin-weave	114 Unidirectional	114-828 Unidirectional
23	Mean = 881 CV = 6.2%	Mean = 918 CV = 9.7%	Mean = 822 CV = 11.2%	Mean = 859 CV = 8.5%	Mean = 2150 CV = 3.5%	Mean = 2018 CV = 8.2%
80	Mean = 822(93) CV = 9.2%	Mean = 888(97) CV = 9.3%	Mean = 778(95) CV = 12.7%	Mean = 829(97) CV = 9.5%	Mean = 1786(83) CV = 7.5%	Mean = 1840(91) CV = 6.3%
130	Mean = 752(85) CV = 11.5%	Mean = 850(93) CV = 6.6%	Mean = 686(83) CV = 14.3	Mean = 715(83) CV = 8.6%	Mean = 1390(65) CV = 6.5%	Mean = 1474(73) CV = 5.3%

Units - MPa, ( ) = % retention of room temperature strength, CV = coefficient of variation

Table 4

THE EFFECT OF INCREASING TEMPERATURE ON THE INTERLAMINAR SHEAR STRENGTH  
OF LAMINATES MADE WITH 114 OR 115 BASED RESIN

Temperature of test °C	114 Satin-weave	114-828 Satin-weave	115 Satin-weave	115-828 Satin-weave	114 Unidirectional	114-828 Unidirectional
23	Mean = 47 CV = 7.8%	Mean = 51 CV = 9.9%	Mean = 42 CV = 6.8%	Mean = 53 CV = 13.3%	Mean = 88 CV = 11.1%	Mean = 92 CV = 5.2%
80	Mean = 44(94) CV = 4.2%	Mean = 47(92) CV = 7.3%	Mean = 36(86) CV = 6.4%	Mean = 39(74) CV = 8.0%	Mean = 65(74) CV = 4.1%	Mean = 71(77) CV = 5.3%
130	Mean = 40(85) CV = 6.0%	Mean = 43(84) CV = 6.6%	Mean = 35(83) CV = 6.3%	Mean = 34(64) CV = 17.5%	Mean = 50(57) CV = 3.9%	Mean = 56(61) CV = 3.7%

Units - MPa, ( ) = % retention of room temperature strength

Table 5  
EFFECT OF IMMERSION IN VARIOUS AIRCRAFT FLUIDS AT 50°C ON  
114-828 UNIDIRECTIONAL COMPOSITE

Fluid	% Weight change 48 h	% Weight change 7 days	Flexural strength* (MPa) 7 days
Methy ethyl ketone	-0.116	-0.044	2166
Avtur 50	-0.076	-0.044	2070
Paint stripper (methylene chloride based)	+0.171	+0.281	2064
Castrol OM 15 (mineral hydraulic oil)	-0.087	-0.079	2081
Skydrol B (phosphate ester based hydraulic oil)	-0.078	-0.016	2227
OX 38 (synthetic lubricating oil)	-0.029	-0.037	2037
Kilfrost ABC (Propylene glycol based de-icer)	+0.143	+0.434	1926
Genklene (1,1,1-trichloroethane)	-0.126	-0.028	2120
Water	+0.169	+0.462	1730

\* Initial flexural strength = 2018 MPa

Fig 1

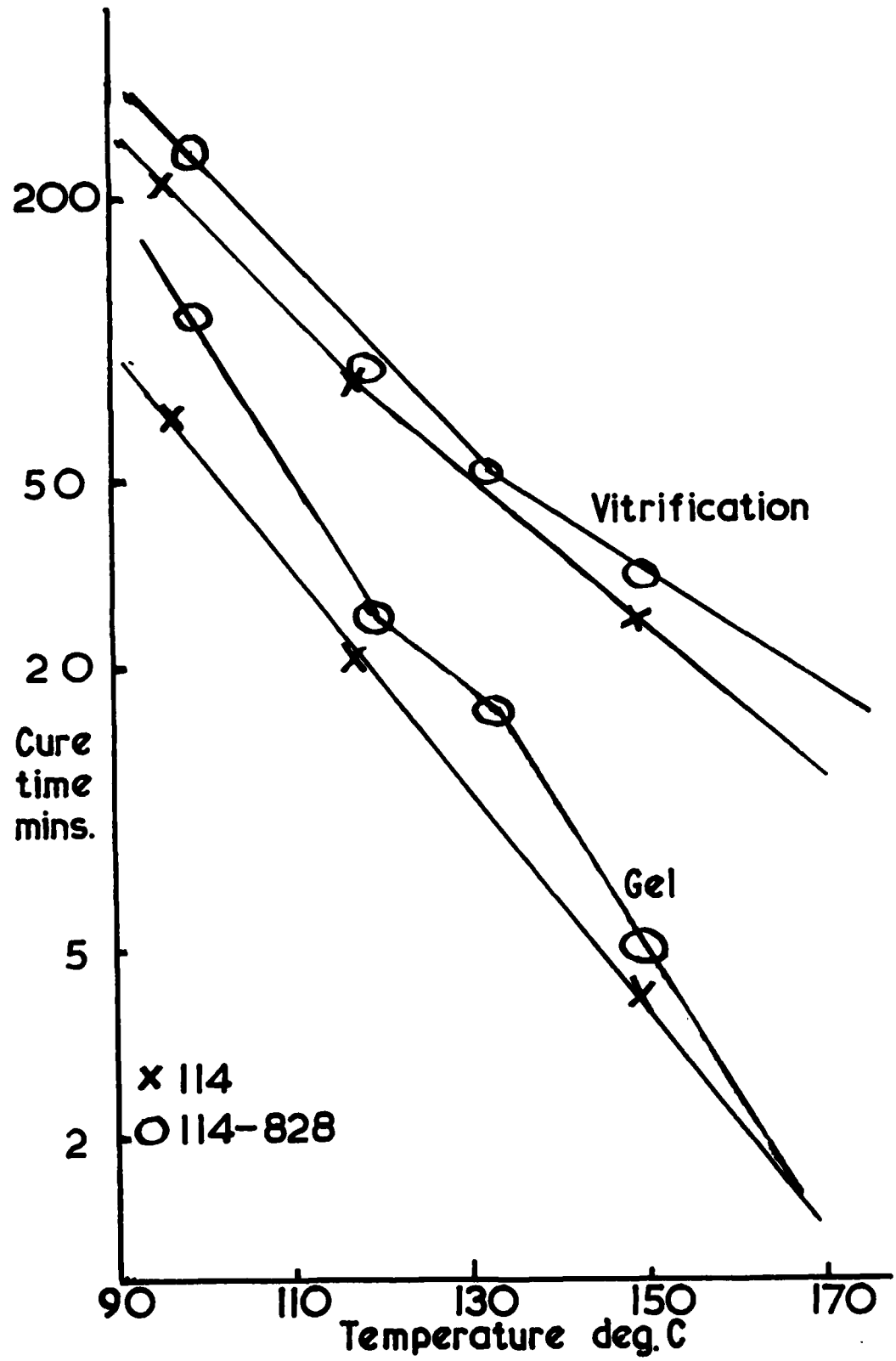


Fig 1 Effect of temperature on the gel and vitrification times of 114 and 114-828 resin systems

Fig 2

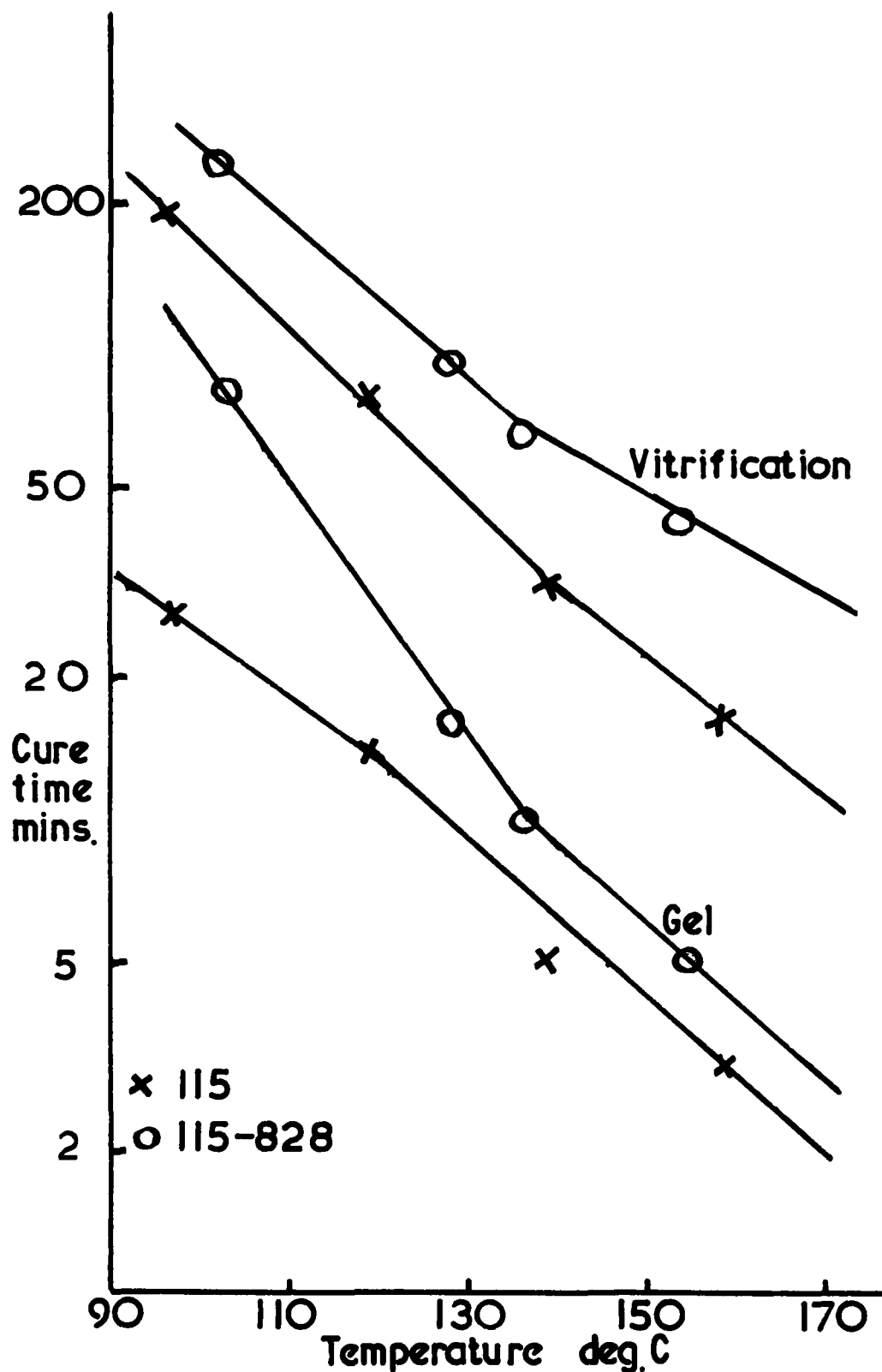


Fig 2 Effect of temperature on the gel and vitrification times of 115 and 115-828 resin systems

**REPORT DOCUMENTATION PAGE**

Overall security classification of this page

**UNCLASSIFIED**

As far as possible this page should contain only unclassified information. If it is necessary to enter classified information, the box above must be marked to indicate the classification, e.g. Restricted, Confidential or Secret.

1. DRIC Reference (to be added by DRIC)	2. Originator's Reference RAE TM Mat 380	3. Agency Reference N/A	4. Report Security Classification/Marking  UNCLASSIFIED
5. DRIC Code for Originator 7673000W	6. Originator (Corporate Author) Name and Location Royal Aircraft Establishment, Farnborough, Hants, UK		
5a. Sponsoring Agency's Code  N/A	6a. Sponsoring Agency (Contract Authority) Name and Location  N/A		
7. Title A new tetra-epoxide resin as a matrix for advanced composites			
7a. (For Translations) Title in Foreign Language			
7b. (For Conference Papers) Title, Place and Date of Conference			
8. Author 1. Surname, Initials Phillips, L.N.	9a. Author 2 Murphy, D.J.	9b. Authors 3, 4 ....	10. Date   Pages   Refs. November   14   8 1981
11. Contract Number N/A	12. Period N/A	13. Project	14. Other Reference Nos.
15. Distribution statement (a) Controlled by – Head of Materials Department, RAE (b) Special limitations (if any) –			
16. Descriptors (Keywords) (Descriptors marked * are selected from TEST) Carbon fibre. Composites. Epoxy resins. Mechanical properties			
17. Abstract  Two versions of a new tetra-epoxide resin were supplied by Shell Laboratories, for evaluation as a possible matrix for carbon fibre.  Because of the absence of tack in the resulting pre-preg both resins were modified at RAE with a minor percentage of the Bisphenol A resin, Shell 828.  All four formulations ( <i>ie</i> modified and unmodified 114 and 115 resins) had their curing characteristics established and were then used in conjunction with Courtaulds XAS carbon fibre, the reinforcement being in the form of satin-weave and unidirectional fabrics.  The mechanical properties of the derived laminates were obtained at room temperature, at 80°C and 130°C. Resistance to a range of aircraft fluids was also examined.  The best results overall were obtained with the lower molecular weight epoxide blended with Bisphenol A resin; and it is recommended that more extensive trials and long-term testing should begin.			

1/15/81

ATE  
LMED  
8-8