

NASA
Technical Memorandum 100960

AVSCOM
Technical Report 88-C-019

AD-A205 573

Surface Fatigue Life of CBN and Vitreous Ground Carburized and Hardened AISI 9310 Spur Gears

Dennis P. Townsend
*Lewis Research Center
Cleveland, Ohio*

and

P.R. Patel
*Bell Helicopter Textron
Fort Worth, Texas*

DTIC
ELECTE
MAR 1 0 1989
H_{co}

Prepared for the
International Off Highway and Powerplant Congress and Exposition
sponsored by the Society of Automatic Engineers
Milwaukee, Wisconsin, September 12-15, 1988

NASA



DISTRIBUTION STATEMENT A

Approved for public release;
Distribution Unlimited

89 2 00 081

SURFACE FATIGUE LIFE OF CBN AND VITREOUS GROUND CARBURIZED
AND HARDENED AISI 9310 SPUR GEARS

Dennis P. Townsend
National Aeronautics and Space Administration
Lewis Research Center
Cleveland, Ohio 44135

and

P.R. Patel
Bell Helicopter Textron
Fort Worth, Texas 76101

ABSTRACT

Spur gear surface endurance tests were conducted to investigate CBN ground AISI 9310 spur gears for use in aircraft applications, to determine their endurance characteristics and to compare the results with the endurance of standard vitreous ground AISI 9310 spur gears. Tests were conducted with VIM-VAR AISI 9310 carburized and hardened gears that were finish ground with either CBN or vitreous grinding methods. Test conditions were an inlet oil temperature of 320 K (116 °F), an outlet oil temperature of 350 K (170 °F), a maximum Hertz stress of 1.71 GPa (248 ksi), and a speed of 10 000 rpm. The CBN ground gears exhibited a surface fatigue life that was slightly better than the vitreous ground gears. The subsurface residual stress of the CBN ground gears was approximately the same as that for the standard vitreous ground gears for the CBN grinding method used.

GRINDING OF CARBURIZED AND HARDENED GEAR TEETH for aircraft application has been standard practice for many years. Grinding is required to produce the required accuracy and surface finish necessary for improved life, reduced noise and dynamic loads for aircraft gears. Until a few years ago, the method used for grinding hardened gears was the standard vitreous grinding wheels. The vitreous grinding method typically produces a very shallow compressive stress (<0.013 mm (0.0005 in.)) on the surface of the ground part but has very little effect on the subsurface compressive residual stress.

A few years ago the cubic boron nitride (CBN) grinding wheels were introduced for grinding gears and other parts (1).^{*} The CBN grinding wheel allows a much greater rate of

stock removal of hardened parts without producing grinding burns that are prevalent with vitreous grinding. The CBN crystals have a high thermal conductivity compared to the vitreous material and conducts the heat away from the part instead of into the part. In addition, the CBN crystals are very sharp and very hard and produce a chip-like cutting action. When a hardened gear or other part is ground very hard with considerable force, a subsurface residual compressive stress is developed below the surface (2). This subsurface residual compressive stress has been shown to improve the subsurface fatigue life of gears and bearings (3,4). The CBN grinding of carburized and hardened AISI 9310 steel spur gears should, therefore, produce equivalent or improved surface fatigue life.

The objectives of the research reported herein were: (1) to investigate CBN grinding as a method for finishing aircraft-type gears, (2) to determine the surface endurance characteristics of CBN ground carburized and hardened AISI 9310 steel spur gears, and (3) to compare the results with standard vitreous ground carburized and hardened AISI 9310 steel spur gears. To accomplish these objectives, tests were conducted with two groups of gears manufactured from one lot of material. One group of spur gears from that lot were CBN ground. For comparison purposes, the other group of spur gears were manufactured by vitreous grinding. The gear pitch diameter was 8.89 cm (3.50 in.) and 3.2 module (8 diametral pitch). Test conditions included an oil inlet temperature of 320 K (116 °F) that resulted in an oil outlet temperature of 350 K (170 °F), a maximum Hertz stress of 1.71 GPa (248 ksi), and a shaft speed of 10 000 rpm.

APPARATUS AND PROCEDURES

GEAR TEST APPARATUS - The gear fatigue tests were performed in the NASA Lewis Research Center's gear fatigue test apparatus (Fig. 1). This test rig uses the four-square principle of applying the test gear load so that the input drive only needs to overcome the frictional losses in the system.

^{*}Numbers in parentheses designate references at end of paper.

A schematic of the test rig is shown in Fig. 1(b). Oil pressure and leakage flow are supplied to the load vanes through a shaft seal. As the oil pressure is increased on the load vanes inside the slave gear, torque is applied to the shaft. This torque is transmitted through the test gears back to the slave gear where an equal but opposite torque is maintained by the oil pressure. This torque on the test gears, which depends on the hydraulic pressure applied to the load vanes, loads the gear teeth to the desired stress level. The two identical test gears can be started under no load, and the load can be applied gradually without changing the running track on the gear teeth.

Separate lubrication systems are provided for the test gears and the main gearbox. The two lubricant systems are separated at the gearbox shafts by pressurized labyrinth seals. Nitrogen is the seal gas. The test gear lubricant is filtered through a 5 μ m nominal fiberglass filter. The test lubricant can be heated electrically with an immersion heater. The skin temperature of the heater is controlled to prevent overheating the test lubricant.

A vibration transducer mounted on the gearbox is used to automatically shut off the test rig when a gear surface fatigue occurs. The gearbox is also automatically shut off if there is a loss of oil flow to either the main gearbox or the test gears, if the test gear oil overheats, or if there is a loss of seal gas pressurization.

The belt-driven test rig can be operated at several fixed speeds by changing pulleys. The operating speed for the tests reported herein was 10 000 rpm.

TEST GEARS - A photograph of the test gears is shown in Fig. 2. The dimensions of the gears are given in Table I. All gears had a nominal surface finish on the tooth face of 0.2 μ m (8 μ in.) rms or better. Typical surface finish charts for both grinding methods are shown in Fig. 3. All gears have a standard 20° involute profile with tip relief. The tip relief was 0.0013 cm (0.0005 in.) starting at the highest point of single tooth contact. One group of gears was ground with a vitreous grinding wheel with speed, feed and metal removal rate as shown in Table II. The second group of gears were ground with a CBN form grinder with speed, feed and metal removal rate as shown in Table II.

Residual stress profiles were established using a gear configuration described in Table III, to determine the difference between the two grinding techniques. For baseline condition, one gear was tested in as-carburized condition. The stress measurements were made using x-ray diffraction technique at the approximate pitch diameter of the gears. The results of residual stress measurements are summarized in Fig. 4.

TEST MATERIAL - The gears were manufactured from vacuum induction melted, vacuum arc remelted (VIM VAR) AISI 9310 steel. The nominal chemical composition of the gears is given in Table IV. The heat treatment procedure for the test gears is given in Table V. The case and core proper-

ties of the test gears are given in Table VI. Photomicrographs of the case and core of a test gear is given in Figs. 5(a) and (b).

TEST LUBRICANT - All the gears were lubricated with a single batch of synthetic paraffinic oil, which was the standard test lubricant for the gear tests. The physical properties of this lubricant are summarized in Table VII. Five percent of an extreme pressure additive, designated Lubrizol 5002 (partial chemical analysis given in Table VII) was added to the lubricant.

TEST PROCEDURE - After the test gears were cleaned to remove their protective coating, they were assembled on the test rig. The test gears were run in an offset condition with a 0.30 cm (0.120 in.) tooth-surface overlap to give a surface load width on the gear face of 0.28 cm (0.110 in.), thereby allowing for an edge radius on the gear teeth. If both faces of the gears were tested, four fatigue tests could be run for each set of gears. All tests were run-in at a load per unit width of 1230 N/cm (700 lb/in.) for 1 hr. The load was then increased to 5800 N/cm (3300 lb/in.), which resulted in a 1.71 GPa (248 ksi) pitch-line maximum Hertz stress. At the pitchline load the tooth bending stress was 0.21 GPa (30 ksi) if plain bending was assumed. However, because there was an offset load, there was an additional stress imposed on the tooth bending stress. Combining the bending and torsional moments gave a maximum stress of 0.26 GPa (37 ksi). This bending stress did not include the effects of tip relief which would also increase the bending stress.

Since the offset test method may introduce edge loading effects the method was originally checked with and without crowned gears. There was no difference between crowned and uncrowned gears, also all fatigue spalls with uncrowned gears originate evenly along the tooth flank and never start at the edge location. This is proof that the offset test condition is an acceptable method for surface fatigue testing.

Operating the test gears at 10 000 rpm gave a pitchline velocity of 46.55 m/sec (9163 ft/min). Lubricant was supplied to the inlet mesh at 800 cm³/min (49 in.³/min) and 320±6 K (116±10 °F). The lubricant outlet temperature was nearly constant at 350±3 K (170±5 °F). The tests ran continuously (24 hr/day) until the rig was automatically shut down by the vibration detection transducer (located on the gearbox adjacent to the test gears) or until 500 hr of operation without failure were completed. The lubricant circulated through 5 μ m fiberglass filter to remove wear particles. For each test, 3.8 liter (1 gal) of lubricant were used. At the end of each test, the lubricant and filter element were discarded. Inlet and outlet oil temperatures were continuously recorded on a strip-chart recorder.

The pitchline elastohydrodynamic (EHD) film thickness was calculated by the method of Ref. 5. It was assumed, for this film thickness calculation, that the gear surface temperature at the pitchline was equal to the outlet oil temperature and that the inlet oil temperature to the contact

even though the oil jet inlet temperature was considerably lower. It is possible that the gear surface temperature was even higher than the oil outlet temperature, especially at the end points of sliding contact. The EHD film thickness for these conditions was computed to be 0.33 μm (13 $\mu\text{in.}$), which gave an initial ratio of film thickness to composite surface roughness (h/s) of 1.15 at the 1.71 GPa (248 ksi) pitchline maximum Hertz stress.

Each pair of gears was considered as a system and, hence, a single test. Test results were evaluated using Weibull plots calculated by the method of Johnson (6). (A Weibull plot is the number of stress cycles versus the statistical percent of gear system failed.)

RESULTS AND DISCUSSION

One lot of VIM VAR AISI 9310 steel spur gears was divided into two groups and endurance tested. One group was ground by a vitreous grinding wheel, while the second group was ground by a CBN form grinding wheel. Test conditions consisted of a tangential tooth load of 5800 N/cm (3300 lb/in.), which produced a maximum Hertz stress of 1.7 GPa (248 ksi), and a speed of 10 000 rpm. The gears failed by classical subsurface pitting fatigue. The pitting fatigue life results of these tests are shown in the Weibull plots of Fig. 6 and are summarized in Table VIII.

Pitting fatigue life results for the gears that were ground by the vitreous grinding wheel are shown in Fig. 6(a). The 10 and 50 percent lines were 82.5×10^6 and 371×10^6 stress cycles (157 and 618 hr), respectively, the Weibull slope was 1.25. The failure index (i.e., the number of fatigue failures out of the number of sets tested) was 6 out of 16. A typical fatigue spall that occurs near the pitchline is shown in Fig. 7. This is a typical fatigue spall similar to those observed in rolling element fatigue tests. The pitchline pitting is the result of a high subsurface shearing stress which develops subsurface cracks. These subsurface cracks propagate into a crack network which results in a fatigue spall that is slightly below the pitchline where the sliding condition is more severe.

Pitting fatigue life results for the gear systems that were ground by a CBN form grinder are shown in Fig. 6(b). The 10 and 50 percent surface fatigue lines were 122.7×10^6 and 502×10^6 stress cycles (205 and 837 hr), respectively, the Weibull slope was 1.34. The failure index was 7 out of 18. The 10 percent surface fatigue life of the CBN ground gears was $\sim 1\frac{1}{2}$ times that of the standard vitreous ground gears. The confidence number was 60 percent, which indicates that there are 600 chances out of 1000 tests that the 10 percent life of the CBN ground gears will be superior to the 10 percent life of the vitreous

ground gears. This indicates that there is not a lot of statistical significance to the life difference between the two groups of gears. However, it does indicate that the CBN gears are at least equivalent in life to the vitreous ground gears or slightly better. The equivalent residual stress profiles of the two methods of grinding would also indicate that the fatigue life should be approximately the same. A more vigorous CBN grinding could induce some additional compressive residual stress; thereby, improving the surface fatigue life, Refs. 2 and 3. A summary of the fatigue lives of the two groups of ground gears are given in Fig. 6(c).

SUMMARY OF RESULTS

Spur gear endurance tests were conducted to investigate CBN ground AISI 9310 spur gears for use in aircraft gear applications, to determine their endurance characteristics and to compare the results with the endurance of standard vitreous ground AISI 9310 spur gears. Tests were conducted with VIM-VAR AISI 9310 carburized and hardened gears that were finish ground with either CBN or vitreous grinding methods. Test conditions were an inlet oil temperature of 320 K (116 °F), an outlet oil temperature of 350 K (170 °F), a maximum Hertz stress of 1.71 GPa (248 ksi), and a speed of 10 000 rpm. The following results were obtained:

1. The CBN ground gears exhibited a surface fatigue life that was slightly better than the vitreous ground gears.
2. The subsurface residual stress of the CBN ground gears was approximately the same as that for the standard vitreous ground gears for the CBN grinding method used.

REFERENCES

1. R.C., DeVries, "Cubic Boron Nitride: Handbook of Properties," REPT-72CRD178, General Electric Co., Schenectady, NY, June 1972. (Avail. NTIS, AD-907330L).
2. G.J., Kimmet, "CBN Finish Grinding of Hardened Spiral Bevel and Hypoid Gears," AGMA Paper 84FTM6, American Gear Manufacturers Association, Alexandria, VA, Oct. 1984.
3. G.A. Johnson, and E. Ratterman, "Enhanced Product Performance Through CBN Grinding Gear Technology, Vol. 5, No. 5, Sept/Oct 1988.
4. H.H., Coe, and E.Z., Zaretsky, "Effect of Interference Fits on Roller Bearing Fatigue Life," ASLE Trans., vol. 30, no. 2, Apr. 1987, pp. 131-140.
5. D., Dowson, and G.R., Higginson, "Elastohydrodynamic Lubrication, The Fundamental of Roller and Gear Lubrication," New York: Pergamon Press, 1966.
6. L.G., Johnson, "The Statistical Treatment of Fatigue Experiments," New York: Elsevier Publication Co., 1964.

A-1

TABLE I. - GEAR DATA

[Gear tolerance per AGMA class 12.]

Number of teeth	28
Diametral pitch	8
Circular pitch, cm (in.)	0.9975 (0.3927)
Whole depth, cm (in.)	0.762 (0.300)
Addendum, cm (in.)	0.318 (0.125)
Chordal tooth thickness reference, cm (in.)	0.485 (0.191)
Pressure angle, deg	20
Pitch diameter, cm (in.)	8.890 (3.500)
Outside diameter, cm (in.)	9.525 (3.750)
Root diameter, cm (in.)	7.988 (3.145)
Root fillet, cm (in.)	0.10 to 0.15 (0.04 to 0.06)
Measurement over pins, cm (in.)	9.603 to 9.630 (3.7807 to 3.7915)
Pin diameter, cm (in.)	0.549 (0.216)
Backlash reference, cm (in.)	0.025 (0.010)
Tip relief, cm (in.)	0.0013 (0.0005)
Tooth width, cm (in.)	0.64 (0.25)

TABLE II. - GRINDING DATA FOR VITREOUS AND CBN GROUND SPUR GEARS

	Wheel speed, rpm	Grit size	Finish μm ($\mu\text{in.}$)	Table speed, sec/pass	Number of passes per tooth	Depth of cut per pass	Time to grind one gear
Vitreous	1600	60	0.36 (14)	2	36	0.018 mm (0.0007 in.)	15 hr
CBN	3400	70	0.30 (12)	6	5	0.13 mm (0.005 in.)	20 min

TABLE III. - DATA FOR GEAR USED FOR
RESIDUAL STRESS MEASUREMENTS

Number of teeth	31
Diametral pitch	8.5
Pressure angle, deg	22
Pitch diameter, cm (in.)	9.264 (3.647)
Face width, cm (in.)	3.386 (1.333)

TABLE IV. - CHEMICAL
COMPOSITION OF TEST
MATERIALS BY
PERCENT WEIGHT

Element	AISI 9310 gears
Carbon (core)	0.10
Manganese	.00
Phosphorus	.006
Sulfur	.005
Silicon	.24
Copper	.04
Chromium	1.35
Molybdenum	.16
Vanadium	.01
Nickel	3.37
Iron	Balance

TABLE V. - HEAT TREAT PROCEDURE FOR TEST GEARS

Pre-carburize heat treatment	
Normalize	1725 °F for 1 hr Air cool
Harden	1500 °F for 1 hr Oil quench
Temper	1000 °F for 4 hr
Carburize	1700 °F for 6.5 hr 1.0 percent carbon potential
Post-carburize heat treatment	
Sub-critical anneal	1150 °F for 2 hr Air cool
Harden	1500 °F for 1 hr Oil quench
Sub-zero treat	-115 °F for 4 hr
Temper	300 °F for 4 hr Air cool

TABLE VI. - CASE AND CORE PROPERTIES OF TEST GEARS

	Conv. Ground	CBN Ground
Surface hardness, HRC	61.5	63.0
HRC 60 depth, mm (in.)	0.45 (0.018)	0.53 (0.021)
HRC 50 depth, mm (in.)	0.99 (0.039)	0.97 (0.038)
Core hardness, HRC	38.0	38.0
Retained austenite, percent	6.0	6.6

TABLE VII. - LUBRICANT PROPERTIES

Property	Synthetic paraffinic oil plus additives ^a
Kinematic viscosity, cm ² /sec (c) at:	
244 K (-20 °F)	2500x10 ⁻² (2500)
311 K (100 °F)	31.6x10 ⁻² (31.6)
372 K (210 °F)	5.5x10 ⁻² (5.5)
477 K (400 °F)	2.0x10 ⁻² (2.0)
Flash point, K (°F)	508 (455)
Fir. point, K (°F)	533 (500)
Pour point, K (°F)	219 (-65)
Specific gravity	0.8285
Vapor pressure at 311 K (100 °F), mm Hg (or torr)	0.1
Specific heat at 311 K (100 °F), J/(kg)(K);(Btu/(lb)(°F))	2190 (0.523)

^aAdditive Lubrizol 5002 (5 vol %): phosphorus 0.03 vol %;
sulfur, 0.93 vol %.

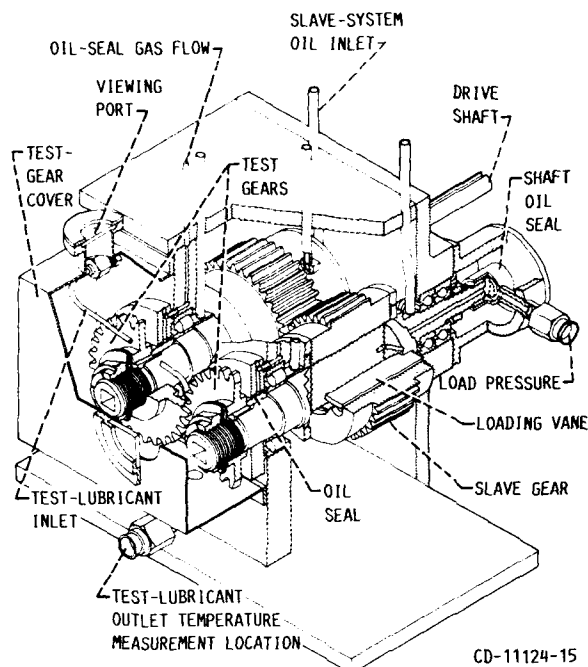
TABLE VIII. - SPUR GEAR FATIGUE LIFE RESULTS

[Pitch diameter, 8.89 cm (3.50 in.); maximum Hertz stress, 1.71 GPa
(248 ksi); speed, 10 000 rpm; lubricant, synthetic paraffinic oil;
gear temperature, 350 K (170 °F).]

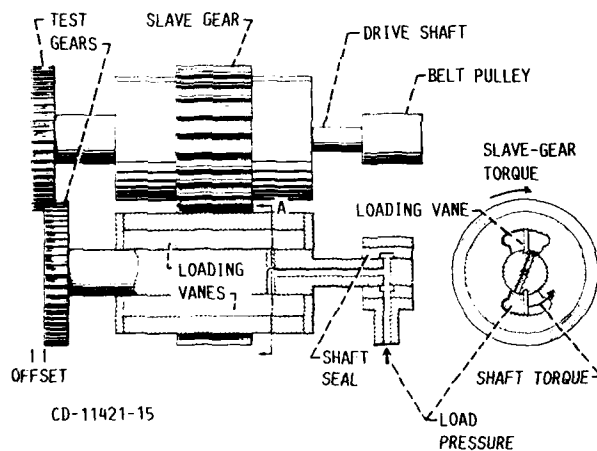
Material	Gear system life, revolutions		Weibull slope	Failure index ^a	Confidence number at 10-percent level ^b
	10-percent life	50-percent life			
Standard ground VIM-VAR AISI 9310	82.5x10 ⁶	371x10 ⁶	1.25	6 out of 16	--
CBW ground VIM-VAR AISI 9310	122.7x10 ⁶	502x10 ⁶	1.34	7 out of 18	60

^aNumber of surface fatigue failures out of number of gears tested.

^bPercentage of time that 10-percent life obtained with AISI 9310 gears
will have the same relation to the 10-percent life obtained with EX-53
gears or CBS 1000 M.



(A) CUTAWAY VIEW.



(B) SCHEMATIC DIAGRAM.

FIGURE 1. - NASA LEWIS RESEARCH CENTER'S GEAR FATIGUE TEST APPARATUS.

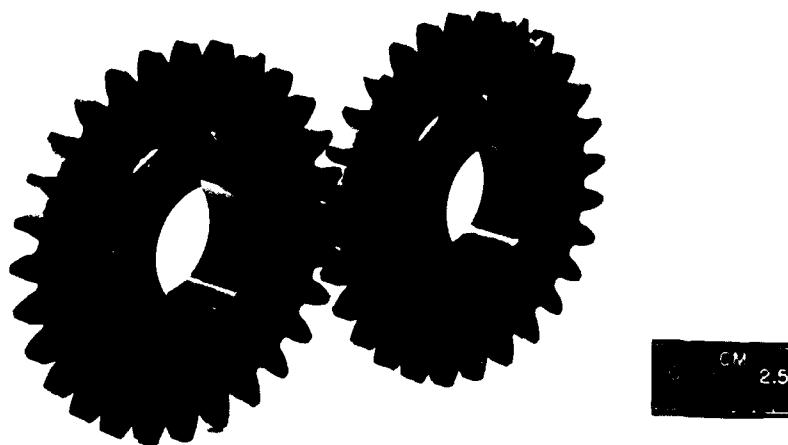


FIGURE 2. - TEST-GEAR CONFIGURATION.

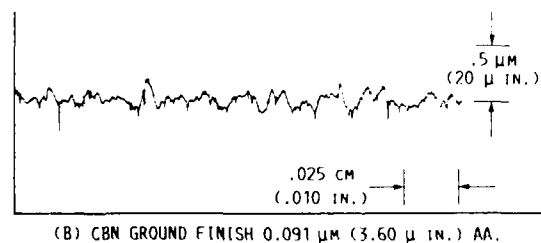
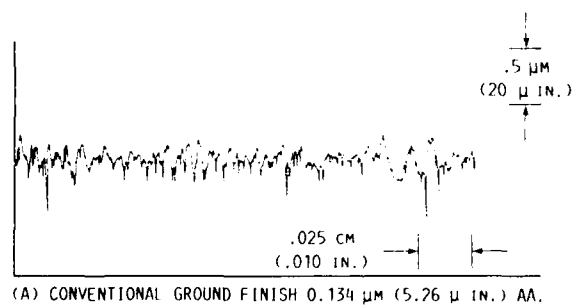


FIGURE 3. - SURFACE FINISH MEASUREMENT IN PROFILE DIRECTION WITH .010 CUTOFF.

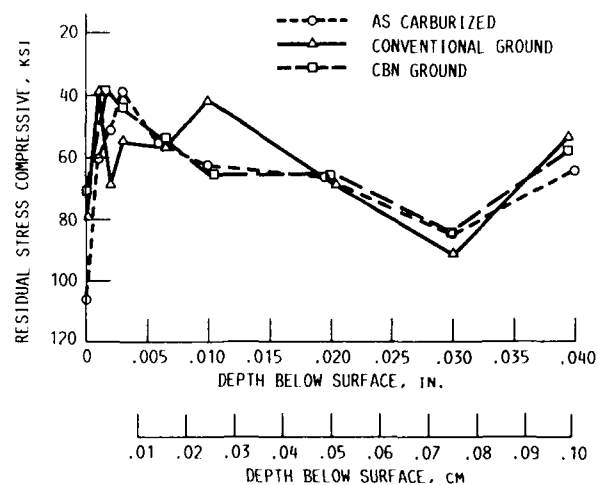


FIGURE 4. - RESIDUAL STRESS MEASUREMENTS ON TOOTH FLANK OF AISI 9310 SPUR GEARS GROUND BY VITREOUS AND CBN GRINDING WHEELS.



(A) CASE.



(B) CORE.

FIGURE 5. - PHOTOMICROGRAPHS OF THE CASE AND CORE MATERIAL FOR TEST GEARS.

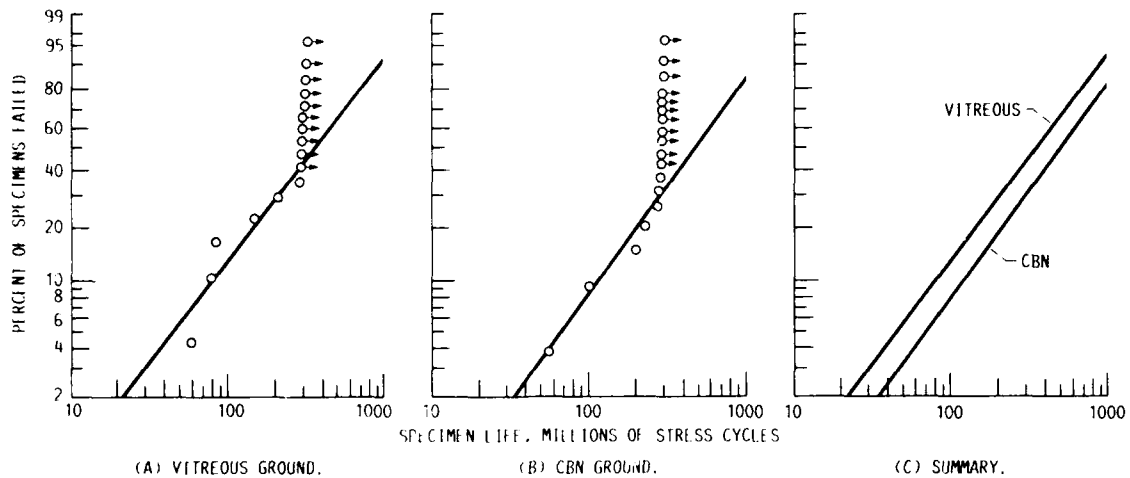


FIGURE 6. - SURFACE FATIGUE LIFE OF CARBURIZED, HARDENED, GROUND AND SHOT PEENED TEST GEARS. SPEED 10 000 RPM; MAXIMUM HERTZ STRESS, 1.71 GPa (.248 KSI); TEMPERATURE, 350 K (170 F); LUBRICANT, SYNTHETIC PARAFFINIC WITH 5-PERCENT E P ADDITIVE.



FIGURE 7. - TYPICAL FATIGUE SPALL FOR AISI 9310 GEARS.



National Aeronautics and
Space Administration

Report Documentation Page

1 Report No. NASA TM-100960 AVSCOM TR-88-C-019	2. Government Accession No.	3. Recipient's Catalog No.
4. Title and Subtitle Surface Fatigue Life of CBN and Vitreous Ground Carburized and Hardened AISI 9310 Spur Gears	5. Report Date	6. Performing Organization Code
7. Author(s) Dennis P. Townsend and P.R. Patel	8. Performing Organization Report No. E-4185	10. Work Unit No. IL162209A47A 505-63-51
9. Performing Organization Name and Address NASA Lewis Research Center Cleveland, Ohio 44135-3191 and Propulsion Directorate U.S. Army Aviation Research and Technology Activity-AVSCOM Cleveland, Ohio 44135-3127	11. Contract or Grant No.	13. Type of Report and Period Covered Technical Memorandum
12. Sponsoring Agency Name and Address National Aeronautics and Space Administration Washington, D.C. 20546-0001 and U.S. Army Aviation Systems Command St. Louis, Mo. 63120-1798	14. Sponsoring Agency Code	
15. Supplementary Notes Prepared for the International Off Highway and Powerplant Congress and Exposition sponsored by the Society of Automotive Engineers, Milwaukee, Wisconsin, September 12-15, 1988. Dennis P. Townsend, NASA Lewis Research Center; P.R. Patel, Bell Helicopter Textron, Fort Worth, Texas 76101.		
16. Abstract Spur gear surface endurance tests were conducted to investigate CBN ground AISI 9310 spur gears for use in aircraft applications, to determine their endurance characteristics and to compare the results with the endurance of standard vitreous ground AISI 9310 spur gears. Tests were conducted with VIM-VAR AISI 9310 carburized and hardened gears that were finish ground with either CBN or vitreous grinding methods. Test conditions were an inlet oil temperature of 320 K (116 °F), an outlet oil temperature of 350 K (170 °F), a maximum Hertz stress of 1.71 GPa (248 ksi), and a speed of 10 000 rpm. The CBN ground gears exhibited a surface fatigue life that was slightly better than the vitreous ground gears. The subsurface residual stress of the CBN ground gears was approximately the same as that for the standard vitreous ground gears for the CBN grinding method used.		
17. Key Words (Suggested by Author(s)) Gears; Surface fatigue life; CBN grinding; Gear life; Residual stress; AISI 9310	18. Distribution Statement Unclassified - Unlimited Subject Category 37	
19. Security Classif. (of this report) Unclassified	20. Security Classif. (of this page) Unclassified	21. No of pages 12
		22. Price* A02