

**NAVAL POSTGRADUATE SCHOOL
MONTEREY, CALIFORNIA**



THESIS

**A STUDY OF FAILURE IN CARBON/FOAM
SANDWICH COMPOSITES WITH STRESS
CONCENTRATION**

by

Peter J. Sistare

September 1995

Thesis Advisor:

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COMPOSITES WITH STRESS CONCENTRATION**

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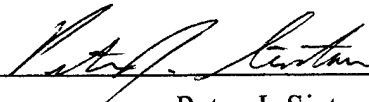
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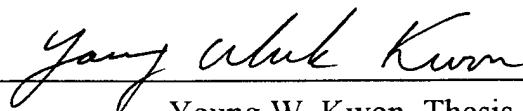
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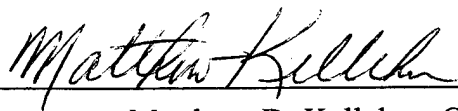


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ABSTRACT

Both experimental and numerical studies were performed to understand the failure mechanism of carbon/foam sandwich composite plates with stress concentration. The plates had circular holes and were subjected to bending and compressive loading. Both three-point and four-point bending tests were conducted. For the testing, the foam thickness, the size of the hole, the number of holes, and the hole location were varied. In addition, a finite element analysis was conducted to verify and understand the experimental results. It was found that four-point bending is not an effective test method to evaluate the effects of stress concentration at a hole. Compressive loading is an effective method. A sample without a hole fails at the quarter point due to foam core shear failure. With a hole at the center, the core shear stress at the quarter point increases with increasing hole size. However, the skin bending stress at the hole increases at a faster rate. When the hole size reaches a critical diameter, the failure mode changes to skin bending failure at the hole.

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I. INTRODUCTION

Sandwich composites are used in the design and construction of various components of today's modern aerospace craft. The reasons behind the selection of these materials are a high strength and stiffness to density ratio, a high resistance to corrosion, and an increased fatigue life. One drawback of this selection is the difficulty in joining components of a craft made of sandwich composites. Mechanical fastening and adhesive bonding are two methods of joining the components. Mechanical fastening allows for relatively easy component replacement. However, the use of bolts or rivets requires the sandwich composite component to be constructed with bolt/ rivet holes. The holes cause stress concentration. The stress concentration can be reduced by the use of buffer strips in the case of laminated composites and by the use of isotropic strain relief inserts in the case of laminates with a high degree of anisotropy. Stress concentration in composite materials around a hole continues to be an area under study. The objective of this study is to further understand the effects of stress concentration on foam cored sandwich composites. The study consists of experimental testing and numerical modeling of a carbon/foam sandwich composite plate with stress concentration due to a circular hole. The plates were subjected to four-point bending, three-point bending, and compressive loading. Foam thickness, hole size, hole location, and the number of holes were varied. A finite element analysis was conducted to verify and understand the experimental results.

II. LITERATURE SURVEY

Lingaiah and Suryanarayana [Ref. 1] conducted an experimental and analytical investigation of the mechanical properties of sandwich composite beams. The beams were constructed of various combinations of fiberglass reinforced plastic and aluminum skins with aluminum honeycomb or foam cores. Four-point and three-point bending tests were conducted. The experimental results showed failure at loads lower than analytically predicted failure loads. Skin-foam bond failure occurred prior to the skin reaching its ultimate tensile/ compressive strength or the foam reaching its ultimate shear stress. The authors also suggested that there was no single source that provided the mechanical properties of all skin/core combinations of a sandwich composite. Therefore the recommendation was that sandwich structures be designed based on experimentally obtained mechanical properties rather than theoretically obtained values.

Clawson [Ref. 2] conducted an experimental investigation of the carbon skin, foam core sandwich composite. The study included impact testing of non-delaminated and delaminated samples and the subsequent residual strength of these samples to withstand a compressive load under simply supported conditions.

Prasad and Shuart [Ref. 3] derived a closed form solution to the moment distribution around a hole in a symmetric composite laminate subjected to a bending moment.

Ueng and Lin [Ref. 4] conducted a numerical study of the stress concentration caused by two identical elliptical holes in composite laminates subjected to in-plane loads and shear loads. Three orthotropic cases were studied. Hole spacing, size, and geometry were varied. It was determined that the effect of a second hole may be ignored if the distance between the hole centers is no more than three to four times the major axis of the elliptical hole.

Meyer and Dharani [Ref. 5] developed an approximate analytical model to determine the stress concentration around a hole in a buffer strip laminate. The results showed that lower modulus and higher failure strain buffer strips are most effective in increasing the strength of laminates with circular cutouts.

Franco and Cloud [Ref. 6] conducted an experimental and analytic study of stress concentration reduction near a hole in an anisotropic glass-epoxy laminate by use of isotropic strain relief inserts. The results showed that the use of stiff strain relief inserts results in the most significant stress concentration reduction.

III. EXPERIMENTAL APPARATUS

The composite beams are of sandwich construction consisting of a rohacell foam core with a thickness of 3.00 mm (0.12 in), 6.35 mm (0.25 in), or 12.70 mm (0.5 in) and two laminated carbon skins. The beam dimensions are 38.10 cm (15 in) long by 3.81 cm (1.5 in) wide. The strain gauges used were Measurements Group Inc type CEA-06-250UN, 350 ohm, gage factor 2.1. Four point bending fixtures were manufactured out of mild steel with the dimensions shown in Figure 1. The fixture contact points are 1/2" diameter steel rods. Three point bending was conducted using the bottom fixture of the four-point apparatus in conjunction with a third point centered above the beam. Column compression testing was conducted using the pinned-pinned end condition fixtures designed and manufactured as described by Clawson [Ref. 2]. The Instron Model 4507 tensile/compression test machine was used with a 200 KN load cell. Load and displacement data was obtained using the Instron Model 4500 data acquisition tower with the Instron series IX automated materials testing software, version 5.28. Strain gauge readings were acquired through the Measurements Group P-3500 strain indicator/SB-10 switch and balance unit combination.

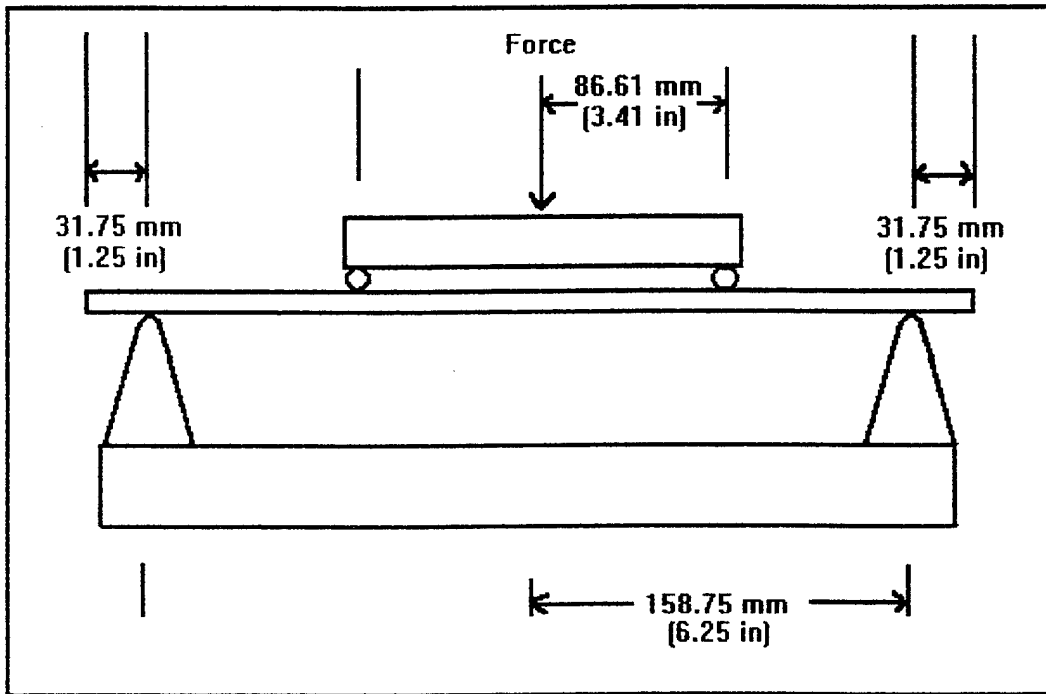


Figure 1. Four Point Bending Fixtures.

IV. BENDING TEST

Four-point and three-point bending tests were conducted to evaluate the effects of stress concentration in a foam core sandwich composite with a hole.

A. FOUR-POINT BENDING

Four-point bending tests were conducted as follows. The first beam had a 6.35 mm (0.25 in.) foam core. Two strain gauges were mounted on the sample as shown in Figure 2.

A cross head speed of 2.54 mm (0.1 in.) per minute was used for all bending tests. The specimen failed at 0.64 KN and a strain of 1350 microns. The results are shown in Figure 3 and Table 1. The point at which the load decreases is considered as the failure point. The failure occurred in the vicinity of the lower support, with delamination and foam core shear failure evident. The 1 to 1 correlation between strain gauge 1 and strain gauge 2 demonstrates symmetry of the upper and lower carbon fiber skins. One skin only will be evaluated when conducting the remaining bending tests. The deformed beam took the shape shown in Figure 4. It is uncharacteristic of classical beam bending and will be discussed further. Figure 5 shows classical beam bending. The next test was conducted under the same conditions but with a 6.35 mm (0.25 in) diameter circular hole drilled at the center of the composite strip. The hole was drilled with a backing strip to avoid delamination of the carbon skin from the foam core. One strain gauge was placed as shown in Figure 6. Failure occurred at a similar load with a higher strain. The failure however was not at the hole but at the lower support as was the case with the no-hole test. See Figure 7 and Table 2. The

deformed shape was also that of Figure 4. A third such test was conducted with a 12.70 mm (0.5 in.) diameter centered hole. The results mimicked those of the first two tests, the failure occurring at the same load and at the lower support. See Figure 8 and Table 3. The deformed beam shape remained the same as the first two tests. The results showed that stress concentration at the support was more critical than that at the hole unless the hole was very large relative to the specimen width.

B. FOUR-POINT AND THREE- POINT CLASSICAL BEAM BENDING

INVESTIGATION

A sample beam was prepared with strain gauge locations as indicated in Figure 9. The beam was then subjected to four point bending and three point bending to determine if the beam deflected according to classical beam bending theory. Referring to Figure 10, classical beam bending theory stipulates that under three point bending, the strain between the lower end support and the upper point load varies linearly with the distance X between them. Similarly, referring to Figure 11, when conducting four-point bending the strain between the lower end support and the upper point load varies linearly with the distance X between them while the strain between the upper two point loads is constant. Four-point bending was conducted three times for repeatability. Strain comparisons of gauges that were not under point loads were within 3 percent when compared to the results predicted by classical beam bending. The strain comparisons between gauges located under point loads corresponded within 42 percent. See Figures 12, 13, and 14 and Tables 4, 5, and 6 respectively for the results. Next, the sample beam was subjected to three point bending within the elastic

region. Again, the test was conducted three times for repeatability. Strain measurements from gauges that were not under point loads agreed with those predicted by classical beam bending theory within 3.2 percent. The strain comparisons between gauges located under point loads corresponded within 4.0 percent. See Figures 15, 16, and 17 and Tables 7, 8, and 9 respectively for the results. The details of the six tests were reviewed to determine why the three-point bending yielded classical beam bending results while the four point bending test did not. It was noted that all load and support points were constructed of steel dowels except the upper point of the three-point bending test which was constructed of a flat plate approximately one inch wide. The results show that the strain gauges located under the dowel points produce higher strains than predicted by classical beam bending while the strain gauges under the flat plates produce strains relatively close to predicted strains. The flat plate acted to reduce the local stress concentration at the point load. Three point bending was conducted again but with the upper point constructed of a dowel. Strain comparisons of the gauge under the upper point deviated from classical beam bending by 33 Percent, up from 4.0 percent with the flat plate. The results are shown in Figure 18 and Table 10.

C. FOUR-POINT AND THREE- POINT BENDING WITH LOCAL STRESS REDUCERS

Flat plates were constructed for all load points and support points of the three and four-point bending fixtures as shown in Figures 19 and 20. The contact point plate widths were 19.05 mm (0.75 in.), 25.4 mm (1.0 in.), and 31.75 mm (1.25 in.). The sample beam was subjected to a series of three point bending tests within the elastic region utilizing the flat

plates at the upper load point and at the two support points. With the 19.05 mm (0.75 in.) plates, all strain readings correlated to predicted strains within 23 percent. When the 25.4 mm (1.0 in.) plates were used, a correlation within 15.5 percent was achieved. The 31.75 mm (1.25 in.) plates resulted in a correlation within 7.6 percent. The test results are shown in Figures 21, 22, and 23 and Tables 11, 12, and 13 respectively. The sample beam was then subjected to four-point bend testing within the elastic region, also utilizing the flat plates at the load points and at the support points. With the 25.4 mm (1.0 in.) plates, all strain readings correlated to predicted strains within 24 percent. When the 31.75 mm (1.25 in.) plates were used, a correlation within 11.82 percent was achieved. See Figures 24 and 25 and Tables 14 and 15 respectively. The data shows that the use of plates at the load point and at the support points when conducting three-point bending resulted in a correlation within 7.6 percent. A correlation within 4 percent was achieved when using a plate at the load point only. An explanation for this difference was not pursued. Three-point bending is not used to test samples with holes. Four-point bending is the test method used for this purpose. When utilized at the load points and at the support points, the plates improved the correlation for the four-point bending and were therefore implemented.

D. FOUR-POINT BENDING TO FAILURE WITH LOCAL STRESS REDUCERS

With the 31.75 mm (1.25 in.) plates acting to reduce the local stress concentrations at all four points, four-point bending was conducted on a no-hole specimen with a core thickness of 6.35 mm (0.25 in.). One strain gauge was mounted at the center of the beam. The beam failed at the lower load point and at a load similar to the failure load obtained when

testing without the plates. See Figure 26 and Table 16. The same test was conducted on a sample with a core thickness of 6.35 mm (0.25 in.) with a 12.70 mm (0.5 in) hole. The beam failed at the lower support point and at a load similar to the failure load obtained when testing this beam without the plates. See Figure 27 and Table 17. Regardless of the utilization of the flat plates, the beam failed in shear at the support point before a critical stress was reached at the hole. Although the flat plate at the support redistributes the skin bending stress around the support, it does not affect core shear stress at the support. As a result, core shear occurred at the support at the same failure load.

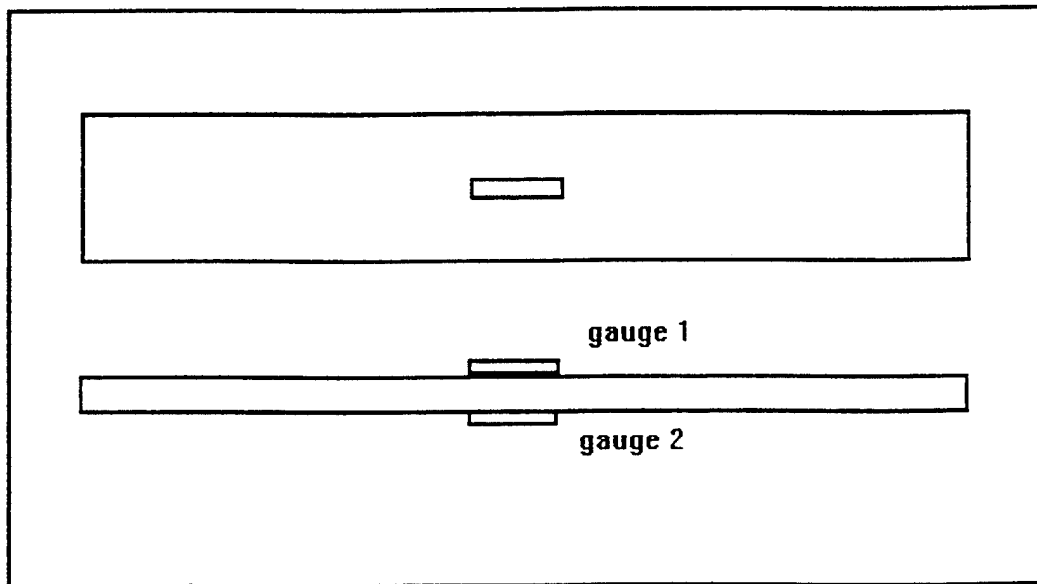


Figure 2. Strain Gauge Locations for Four Point Bending, No Hole.

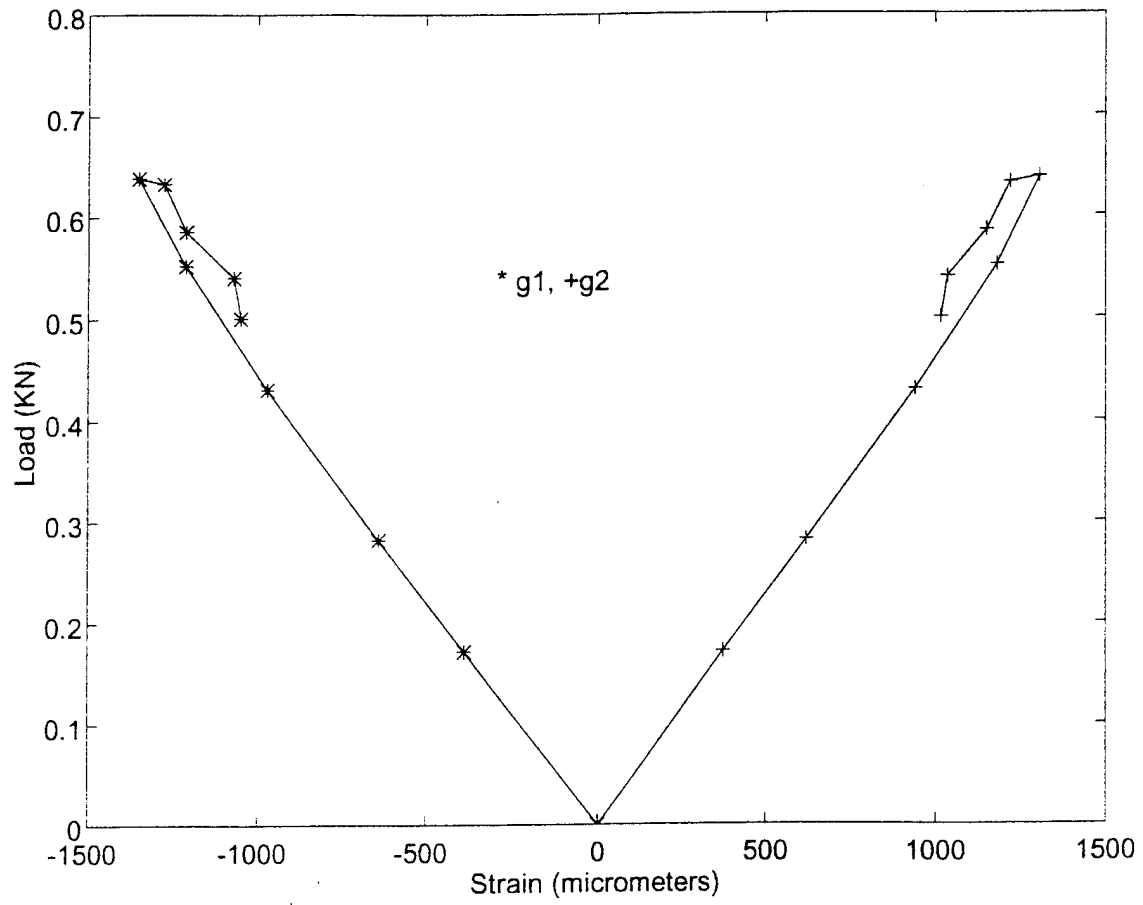


Figure 3. Load- Strain Diagram for 6.35 mm Thick, No Hole.

| load (KN) | gauge1 | gauge2 |
|-----------|--------|--------|
| 0.17 | -384 | 375 |
| 0.28 | -639 | 620 |
| 0.43 | -970 | 943 |
| 0.55 | -1212 | 1180 |
| 0.64 | -1350 | 1306 |
| 0.63 | -1275 | 1220 |
| 0.59 | -1210 | 1150 |
| 0.54 | -1070 | 1036 |
| 0.50 | -1050 | 1017 |

Table 1. Results for Four Point Bending, No Hole.

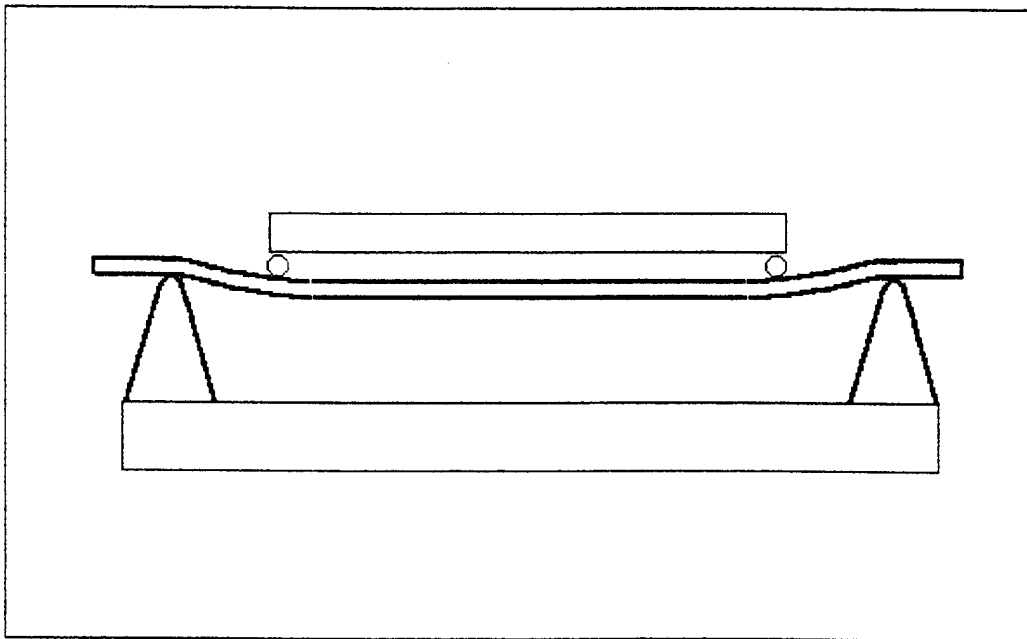


Figure 4. Bending Uncharacteristic of Classical Beam Bending Theory.

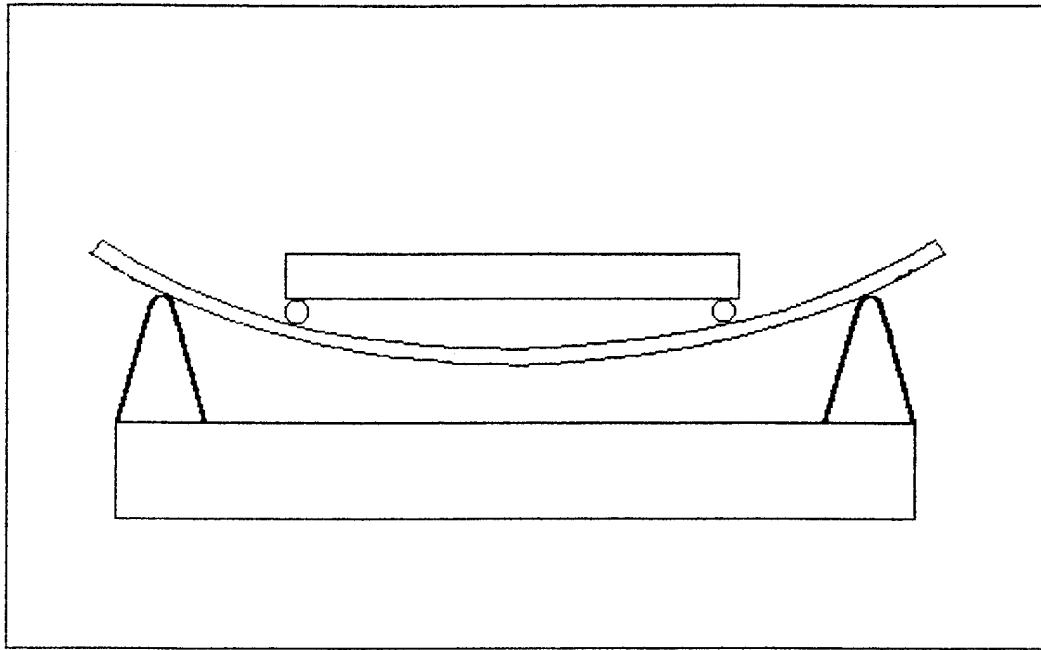


Figure 5. Bending Characteristic of Classical Beam Bending Theory.

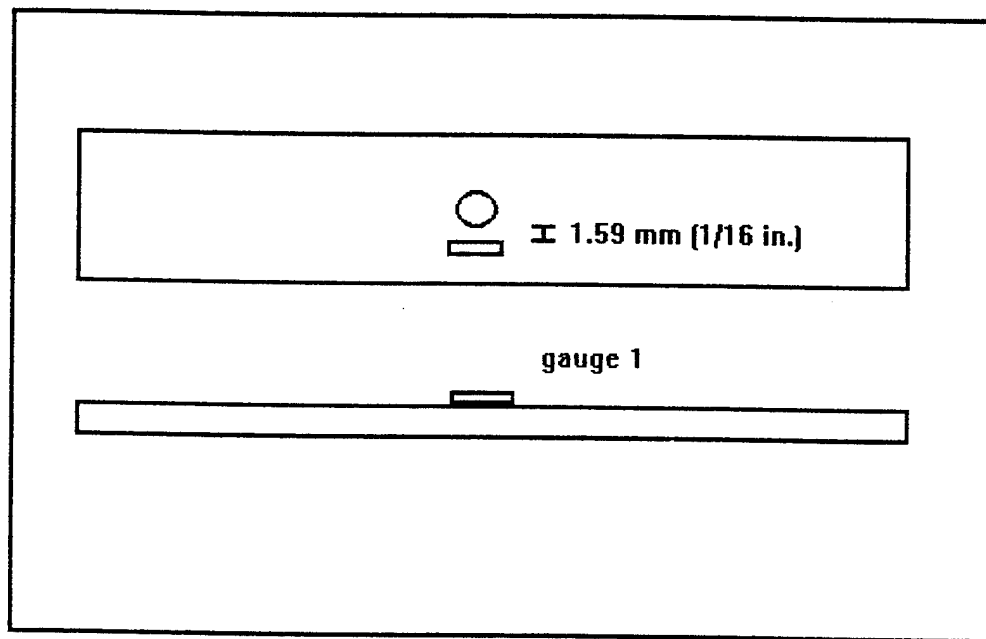


Figure 6. Strain Gauge Location for Four Point Bending, One 6.35 mm Hole.

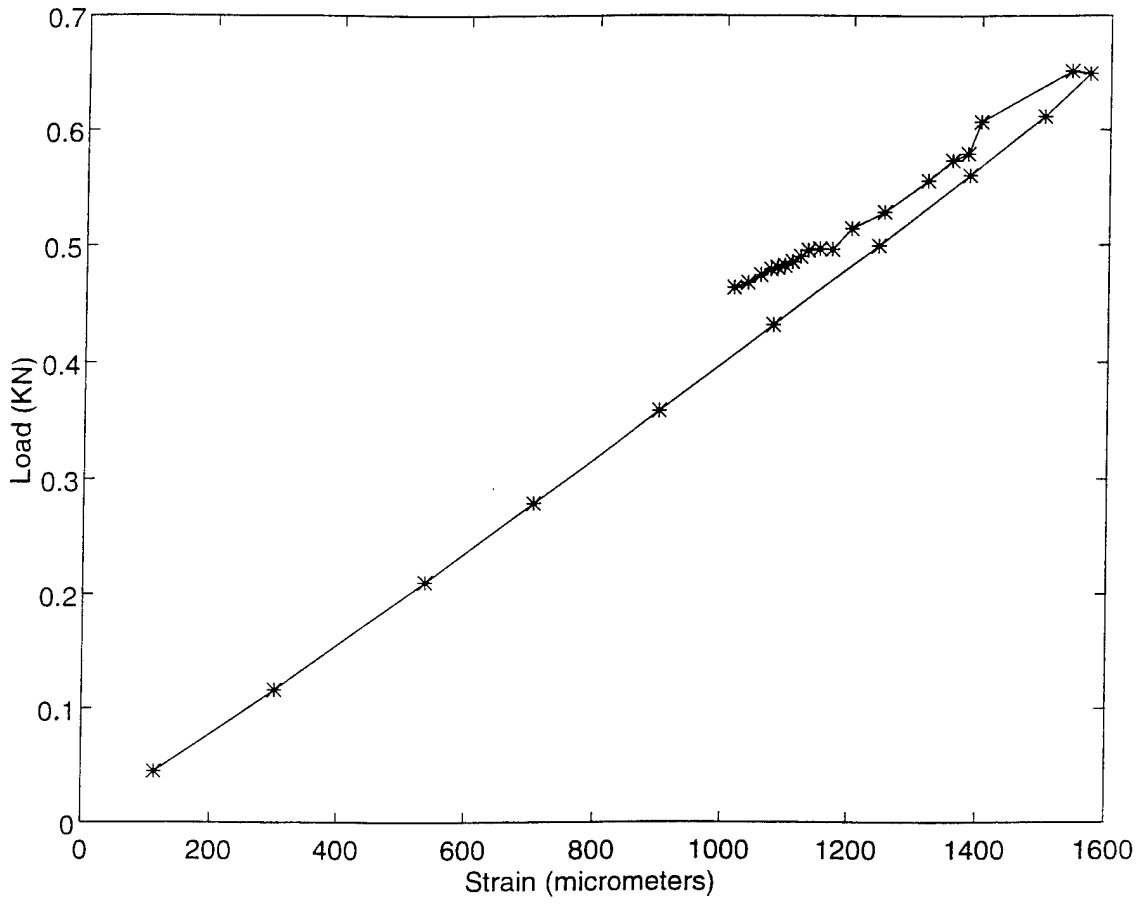


Figure 7. Load-Strain Diagram for 6.35 mm Thick, One 6.35mm Hole.

| load (KN) | gauge1 |
|-----------|--------|
| 0.04 | 113 |
| 0.12 | 301 |
| 0.21 | 536 |
| 0.28 | 705 |
| 0.36 | 901 |
| 0.43 | 1078 |
| 0.50 | 1242 |
| 0.56 | 1383 |
| 0.61 | 1499 |
| 0.65 | 1568 |
| 0.65 | 1540 |
| 0.61 | 1400 |
| 0.58 | 1380 |
| 0.57 | 1356 |
| 0.56 | 1318 |
| 0.53 | 1250 |
| 0.51 | 1200 |
| 0.50 | 1170 |
| 0.50 | 1150 |
| 0.50 | 1132 |
| 0.49 | 1120 |
| 0.49 | 1107 |
| 0.48 | 1095 |
| 0.48 | 1083 |
| 0.48 | 1073 |
| 0.48 | 1057 |
| 0.47 | 1037 |
| 0.46 | 1015 |

Table 2. Results for Four Point Bending, One 6.35 mm Hole.

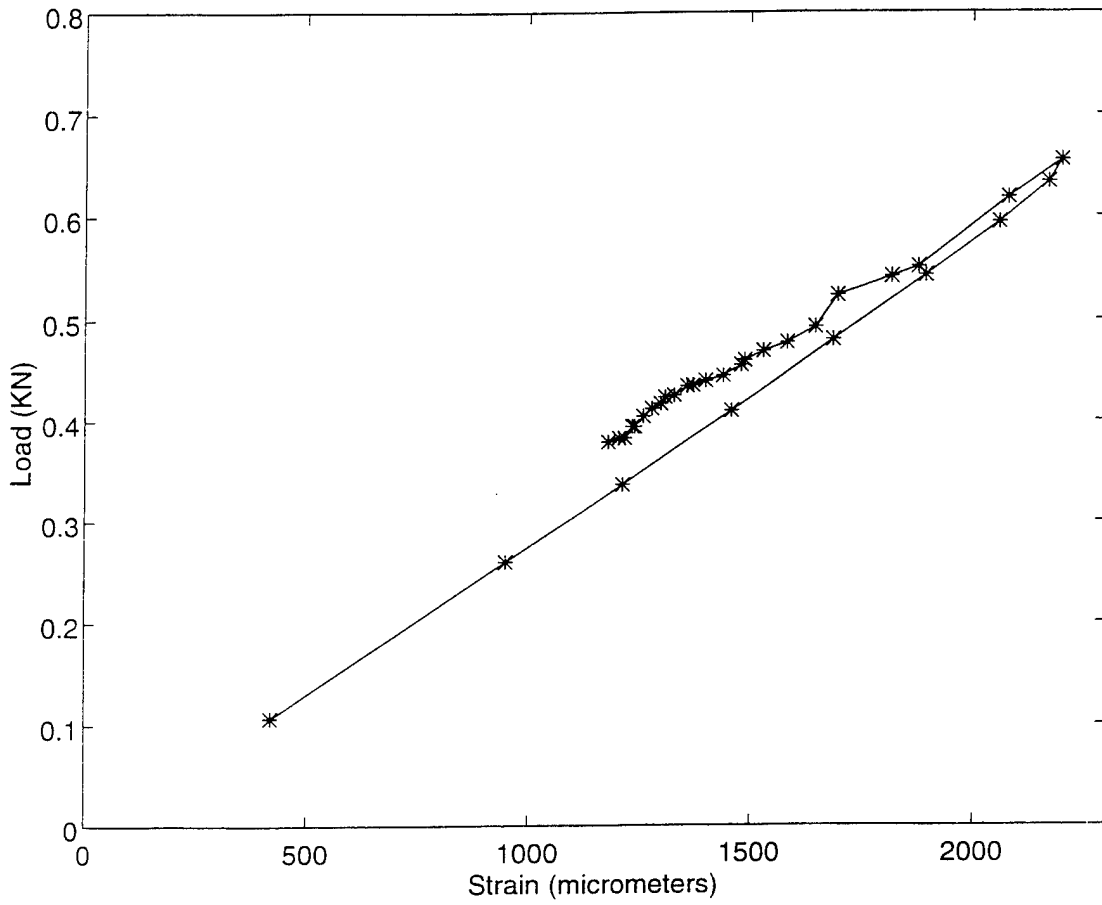


Figure 8. Load-Strain Diagram for 6.35 mm Thick, One 12.70 mm Hole.

| load (KN) | gauge1 |
|-----------|--------|
| 0.11 | 420 |
| 0.26 | 950 |
| 0.34 | 1213 |
| 0.41 | 1458 |
| 0.48 | 1690 |
| 0.54 | 1896 |
| 0.59 | 2060 |
| 0.63 | 2170 |
| 0.65 | 2200 |
| 0.62 | 2080 |
| 0.55 | 1880 |
| 0.54 | 1820 |
| 0.52 | 1700 |
| 0.49 | 1650 |
| 0.48 | 1585 |
| 0.47 | 1530 |
| 0.46 | 1488 |
| 0.46 | 1480 |
| 0.45 | 1440 |
| 0.44 | 1400 |
| 0.44 | 1372 |
| 0.44 | 1360 |
| 0.43 | 1330 |
| 0.42 | 1310 |
| 0.42 | 1300 |
| 0.41 | 1280 |
| 0.41 | 1260 |
| 0.40 | 1240 |
| 0.40 | 1235 |
| 0.38 | 1218 |
| 0.38 | 1205 |
| 0.38 | 1180 |

Table 3. Results for Four Point Bending, One 12.70 mm Hole.

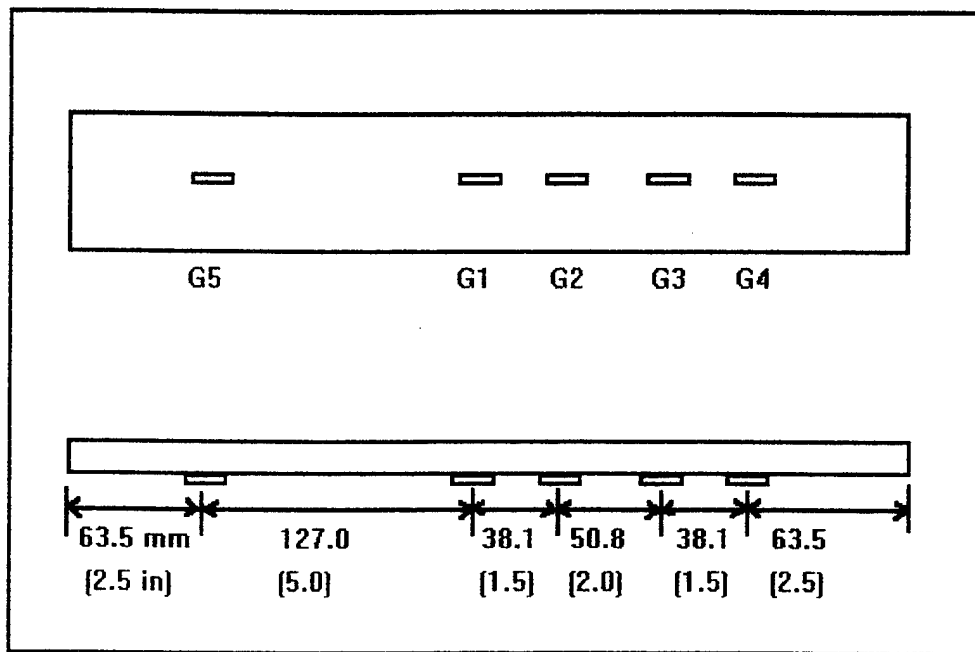


Figure 9. Strain Gauge Locations for Classical Beam Bending Investigation.

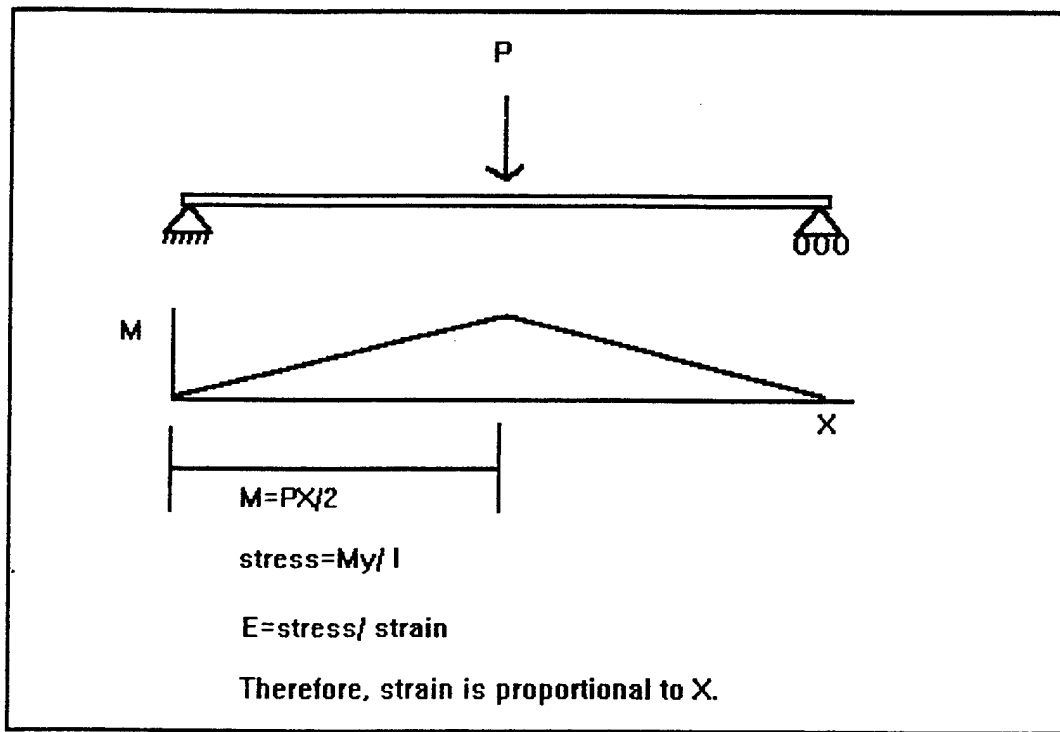


Figure 10. Classical Beam Bending Theory, Three Point Bending.

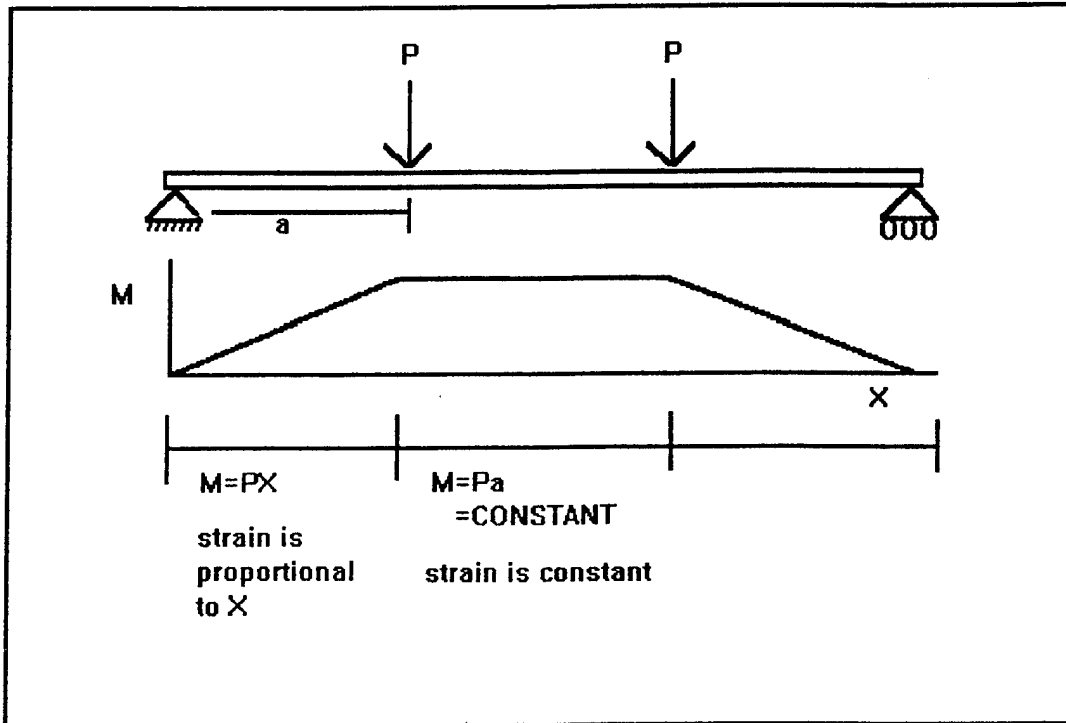


Figure 11. Classical Beam Bending Theory, Four Point Bending.

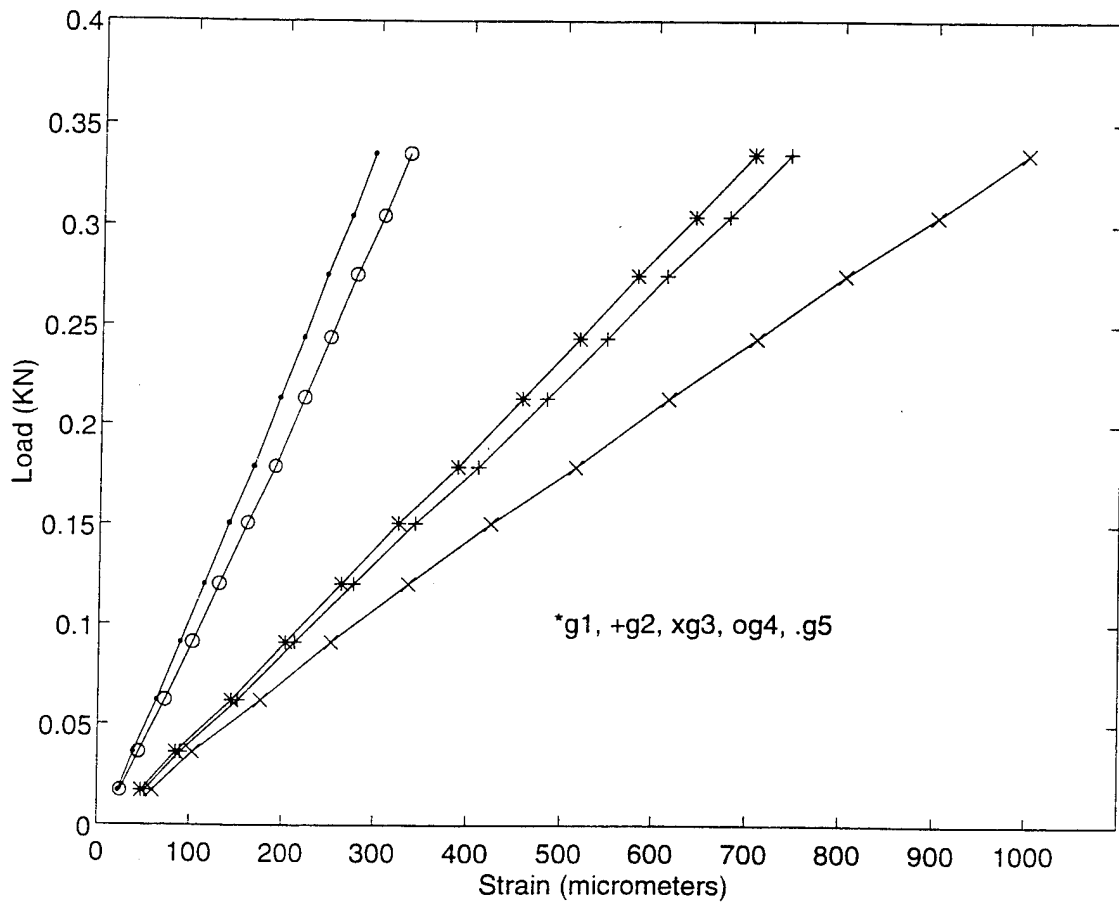


Figure 12. Classical Beam Bending Investigation, Four Point Bending Run 1.
 (See Figure 9 for strain gauge locations.)

| load (KN) | gauge1 | gauge2 | gauge3 | gauge4 | gauge5 |
|-----------|--------|--------|--------|--------|--------|
| 0.02 | 48 | 52 | 60 | 25 | 23 |
| 0.04 | 85 | 90 | 103 | 45 | 39 |
| 0.06 | 144 | 151 | 176 | 73 | 64 |
| 0.09 | 203 | 213 | 253 | 102 | 89 |
| 0.12 | 264 | 277 | 336 | 130 | 114 |
| 0.15 | 325 | 343 | 424 | 160 | 140 |
| 0.18 | 388 | 410 | 514 | 190 | 166 |
| 0.21 | 456 | 482 | 615 | 221 | 194 |
| 0.24 | 517 | 547 | 708 | 249 | 22 |
| 0.28 | 580 | 612 | 803 | 277 | 245 |
| 0.30 | 642 | 678 | 901 | 306 | 271 |
| 0.34 | 704 | 743 | 999 | 333 | 295 |

Table 4. Results for Classical Beam Bending Investigation, Four Point Bending Run 1.

| load (KN) | gauge1 | gauge2 | gauge3 | gauge4 | gauge5 |
|-----------|--------|--------|--------|--------|--------|
| 0.01 | 46 | 50 | 54 | 24 | 22 |
| 0.03 | 89 | 94 | 105 | 46 | 41 |
| 0.06 | 148 | 156 | 178 | 74 | 67 |
| 0.09 | 212 | 225 | 265 | 106 | 93 |
| 0.12 | 274 | 290 | 351 | 134 | 117 |
| 0.16 | 349 | 370 | 458 | 169 | 147 |
| 0.19 | 412 | 438 | 552 | 199 | 173 |
| 0.22 | 476 | 506 | 651 | 229 | 199 |
| 0.25 | 547 | 582 | 761 | 262 | 228 |
| 0.29 | 612 | 650 | 863 | 292 | 255 |
| 0.32 | 673 | 714 | 959 | 320 | 280 |
| 0.35 | 738 | 782 | 1062 | 348 | 307 |

Table 5. Results for Classical Beam Bending Investigation, Four Point Bending Run 2.

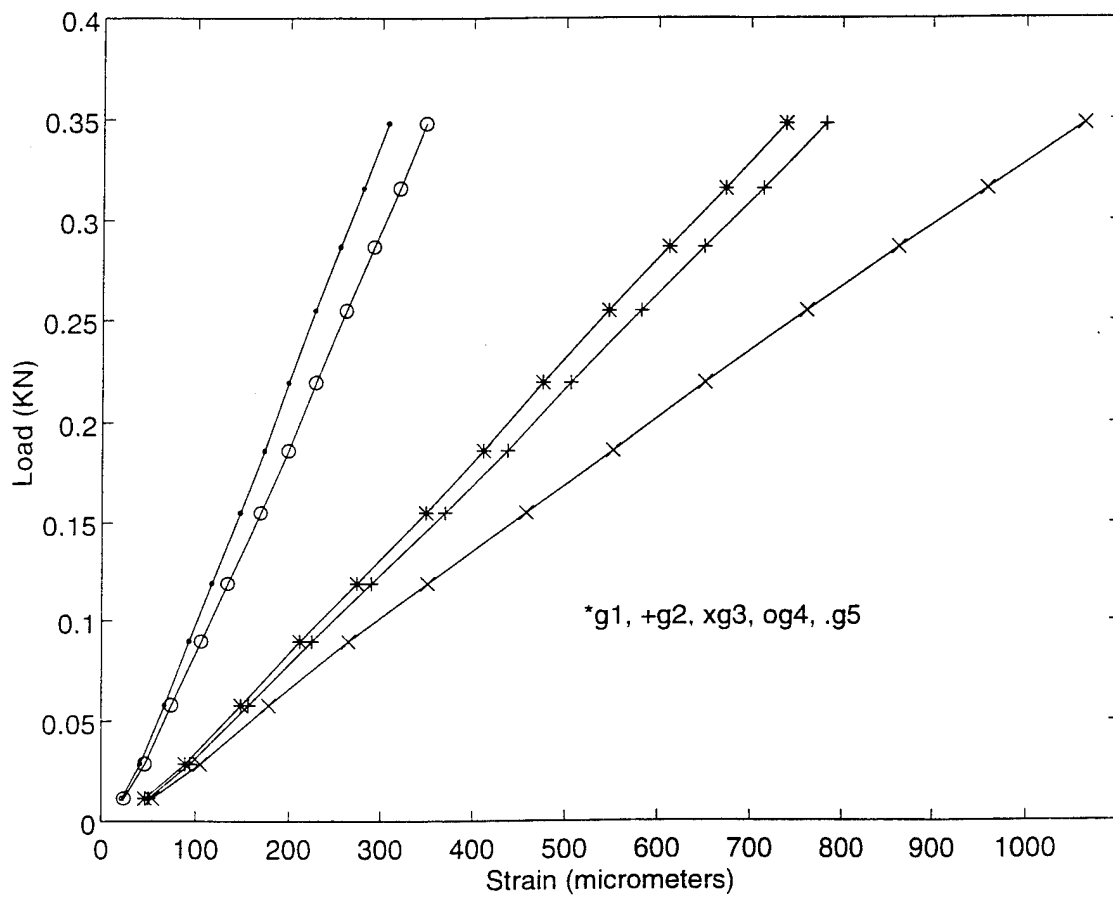


Figure 13. Classical Beam Bending Investigation, Four Point Bending Run 2.
 (See Figure 9 for strain gauge locations.)

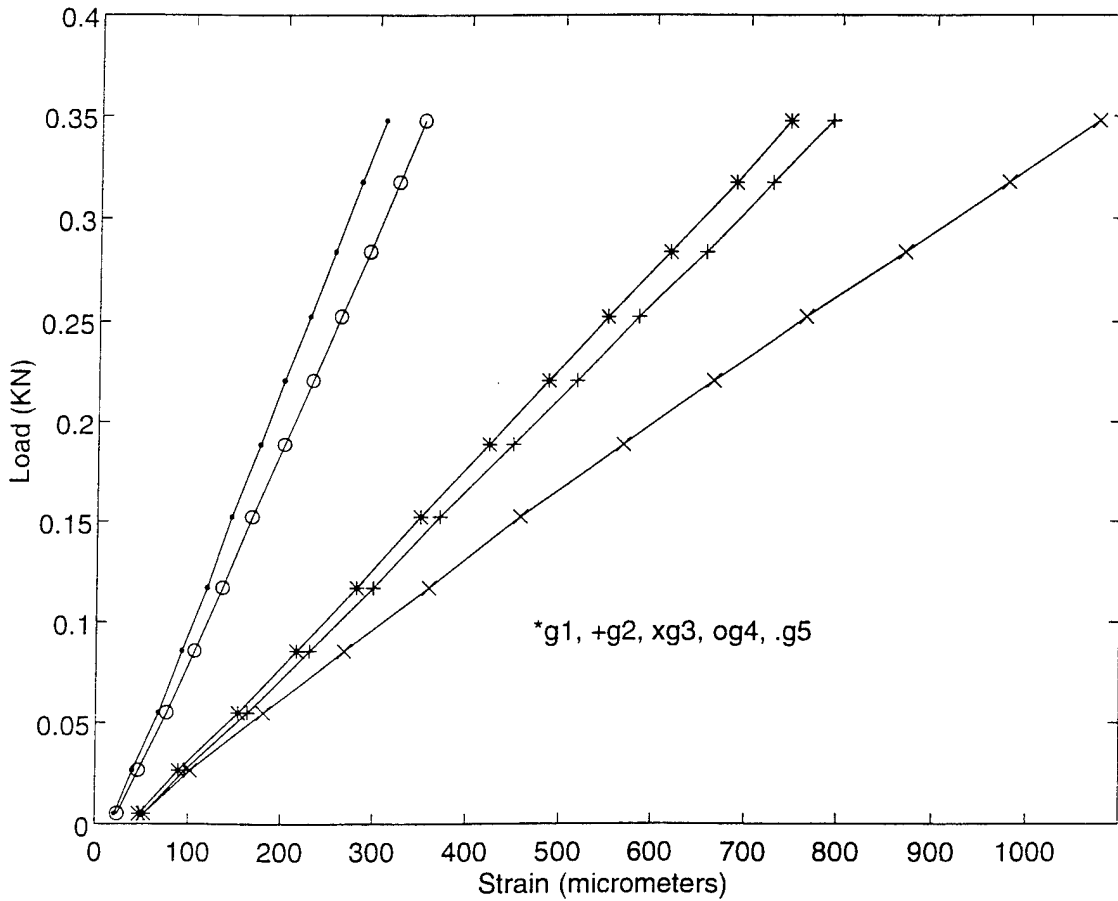


Figure 14. Classical Beam Bending Investigation, Four Point Bending Run 3.
 (See Figure 9 for strain gauge locations.)

| load (KN) | gauge1 | gauge2 | gauge3 | gauge4 | gauge5 |
|-----------|--------|--------|--------|--------|--------|
| 0.01 | 47 | 52 | 52 | 24 | 21 |
| 0.03 | 89 | 96 | 102 | 46 | 40 |
| 0.06 | 154 | 164 | 181 | 76 | 67 |
| 0.09 | 216 | 230 | 268 | 106 | 92 |
| 0.12 | 281 | 299 | 359 | 136 | 119 |
| 0.15 | 349 | 370 | 457 | 167 | 145 |
| 0.19 | 423 | 449 | 567 | 201 | 175 |
| 0.22 | 486 | 516 | 664 | 231 | 200 |
| 0.25 | 549 | 583 | 764 | 261 | 227 |
| 0.28 | 617 | 655 | 868 | 292 | 254 |
| 0.32 | 686 | 726 | 977 | 323 | 282 |
| 0.35 | 745 | 791 | 1074 | 350 | 308 |

Table 6. Results for Classical Beam Bending Investigation, Four Point Bending Run 3.

| load (KN) | gauge1 | gauge2 | gauge3 | gauge4 | gauge5 |
|-----------|--------|--------|--------|--------|--------|
| 0.01 | 57 | 44 | 23 | 14 | 11 |
| 0.03 | 122 | 91 | 51 | 26 | 22 |
| 0.04 | 188 | 140 | 79 | 40 | 33 |
| 0.05 | 250 | 186 | 105 | 52 | 44 |
| 0.06 | 316 | 234 | 133 | 65 | 56 |
| 0.08 | 379 | 281 | 158 | 77 | 66 |
| 0.09 | 452 | 334 | 189 | 90 | 78 |
| 0.10 | 527 | 391 | 221 | 105 | 90 |
| 0.12 | 607 | 450 | 254 | 118 | 103 |
| 0.13 | 680 | 503 | 284 | 130 | 115 |
| 0.15 | 758 | 560 | 317 | 146 | 128 |
| 0.16 | 831 | 615 | 347 | 160 | 140 |
| 0.18 | 906 | 672 | 381 | 173 | 153 |
| 0.19 | 973 | 724 | 411 | 186 | 165 |
| 0.21 | 1044 | 780 | 442 | 200 | 178 |
| 0.22 | 1118 | 838 | 475 | 214 | 191 |
| 0.24 | 1183 | 888 | 504 | 227 | 204 |
| 0.25 | 1260 | 948 | 538 | 243 | 219 |
| 0.27 | 1331 | 1005 | 570 | 257 | 233 |
| 0.28 | 1401 | 1059 | 601 | 270 | 247 |

Table 7. Results for Classical Beam Bending Investigation, Three Point Bending Run 1.

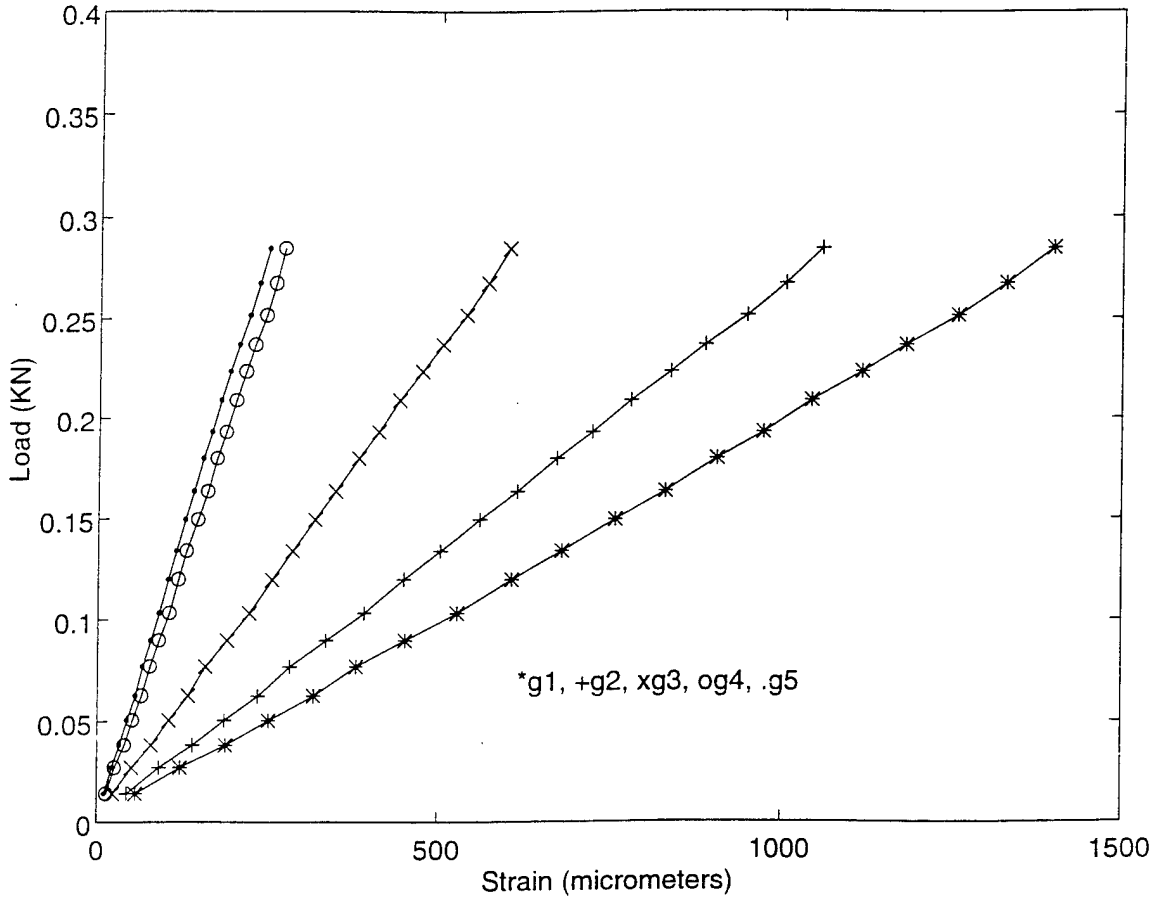


Figure 15. Classical Beam Bending Investigation, Three Point Bending Run 1.
 (See Figure 9 for strain gauge locations.)

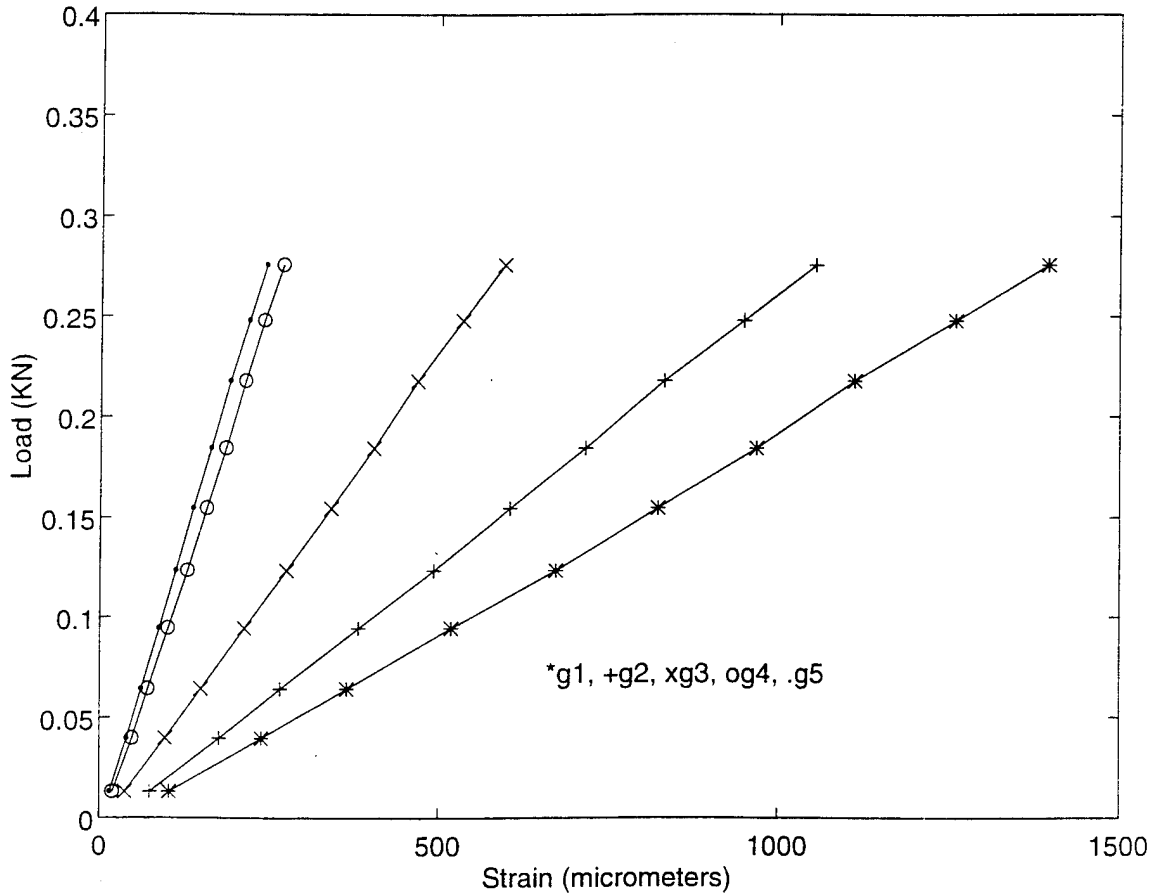


Figure 16. Classical Beam Bending Investigation, Three Point Bending Run 2.
 (See Figure 9 for strain gauge locations.)

| load (KN) | gauge1 | gauge2 | gauge3 | gauge4 | gauge5 |
|-----------|--------|--------|--------|--------|--------|
| 0.01 | 101 | 73 | 37 | 19 | 15 |
| 0.04 | 237 | 174 | 95 | 47 | 39 |
| 0.06 | 363 | 265 | 147 | 70 | 60 |
| 0.09 | 519 | 380 | 211 | 99 | 86 |
| 0.12 | 673 | 493 | 274 | 127 | 110 |
| 0.15 | 823 | 605 | 339 | 155 | 135 |
| 0.18 | 968 | 716 | 402 | 183 | 161 |
| 0.22 | 1114 | 832 | 468 | 212 | 189 |
| 0.25 | 1261 | 949 | 535 | 240 | 217 |
| 0.28 | 1396 | 1056 | 596 | 268 | 243 |

Table 8. Results for Classical Beam Bending Investigation, Three Point Bending Run 2.

| load (KN) | gauge1 | gauge2 | gauge3 | gauge4 | gauge5 |
|-----------|--------|--------|--------|--------|--------|
| 0.01 | 96 | 68 | 33 | 17 | 13 |
| 0.03 | 232 | 169 | 91 | 44 | 37 |
| 0.06 | 360 | 263 | 144 | 69 | 58 |
| 0.09 | 515 | 376 | 208 | 97 | 83 |
| 0.12 | 667 | 487 | 270 | 124 | 108 |
| 0.15 | 824 | 604 | 337 | 154 | 134 |
| 0.18 | 968 | 717 | 401 | 181 | 160 |
| 0.21 | 1112 | 832 | 467 | 210 | 187 |
| 0.24 | 1260 | 950 | 534 | 241 | 216 |
| 0.28 | 1396 | 1058 | 594 | 267 | 242 |

Table 9. Results for Classical Beam Bending Investigation, Three Point Bending Run 3.

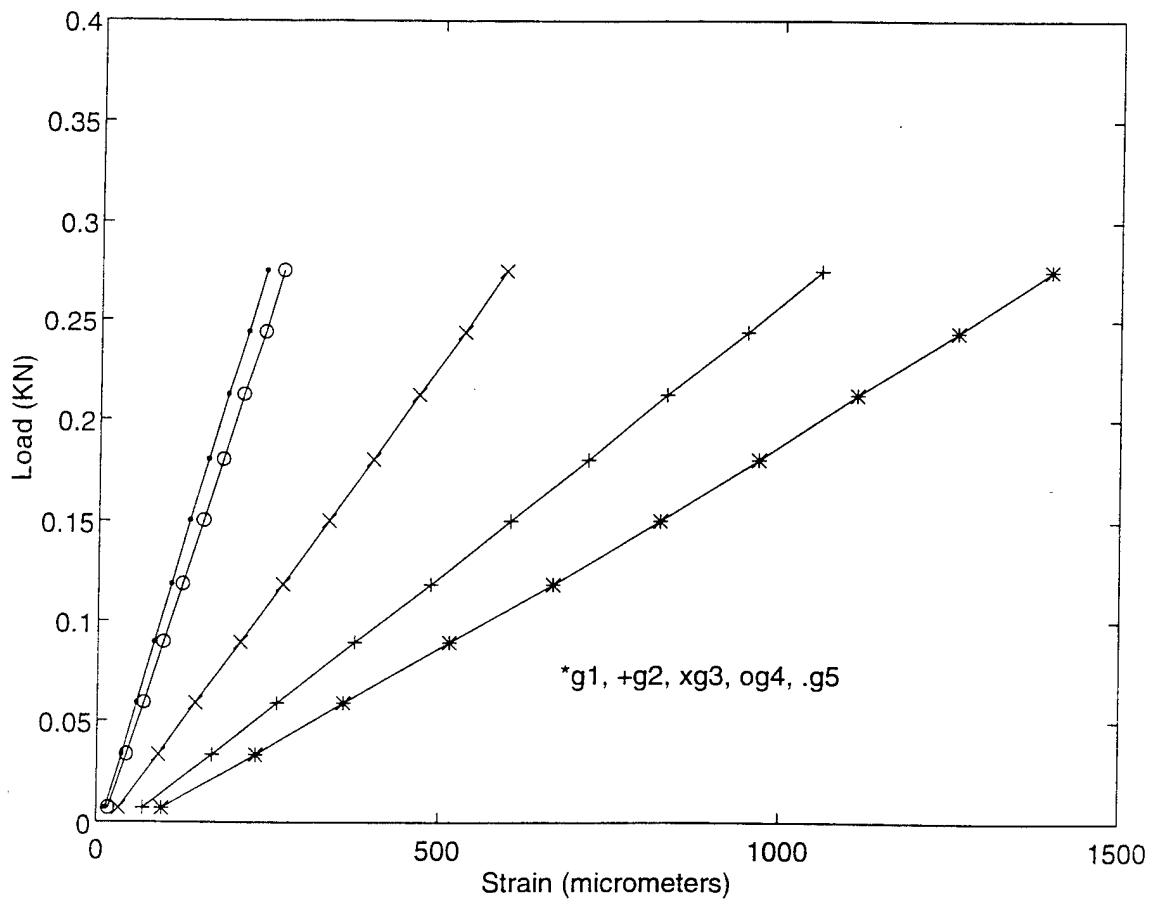


Figure 17. Classical Beam Bending Investigation, Three Point Bending Run 3.
 (See Figure 9 for strain gauge locations.)

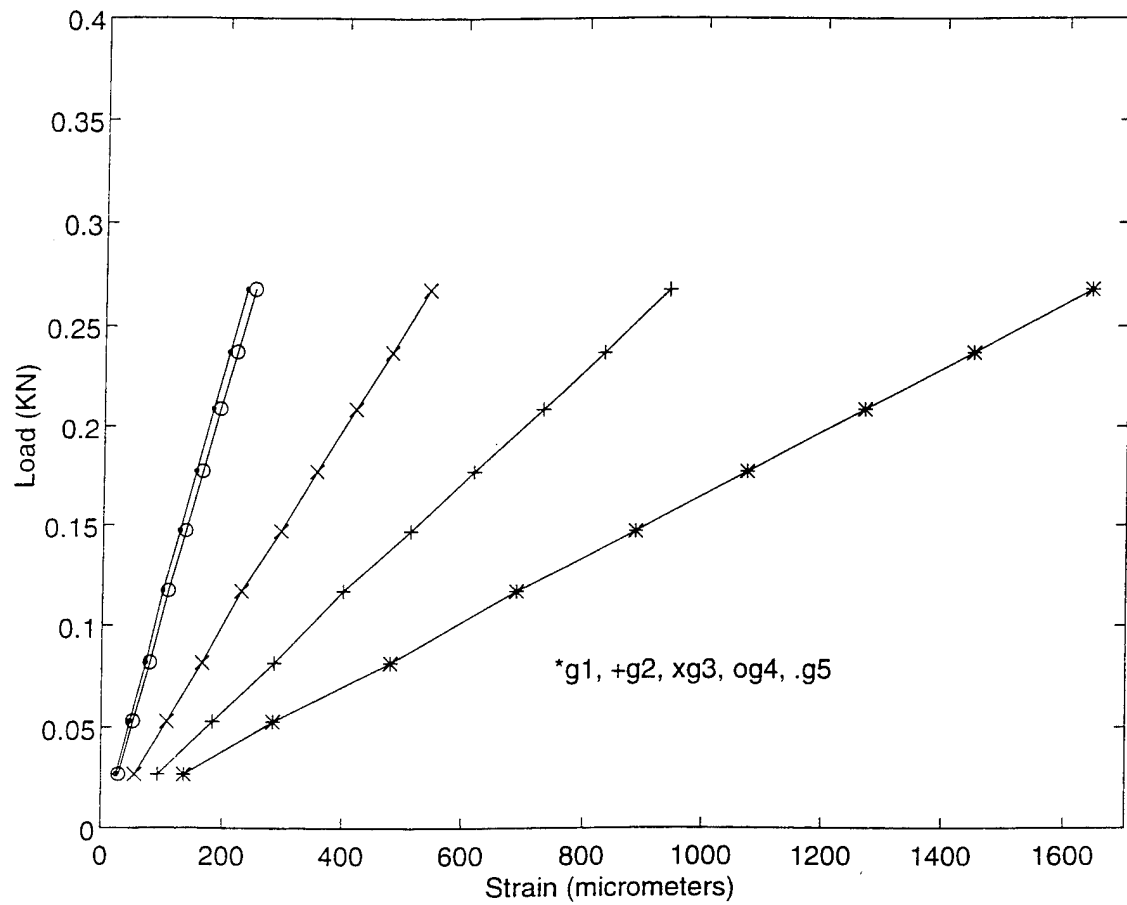


Figure 18. Three Point Bending With Flat Upper Point.
 (See Figure 9 for strain gauge locations.)

| load (KN) | gauge1 | gauge2 | gauge3 | gauge4 | gauge5 |
|-----------|--------|--------|--------|--------|--------|
| 0.03 | 136 | 93 | 55 | 29 | 25 |
| 0.05 | 284 | 183 | 107 | 52 | 47 |
| 0.08 | 480 | 285 | 165 | 79 | 72 |
| 0.12 | 688 | 400 | 229 | 108 | 99 |
| 0.15 | 887 | 512 | 294 | 136 | 126 |
| 0.18 | 1071 | 617 | 353 | 163 | 152 |
| 0.21 | 1268 | 729 | 418 | 191 | 179 |
| 0.24 | 1449 | 831 | 478 | 218 | 205 |
| 0.27 | 1641 | 940 | 540 | 247 | 233 |

Table 10. Results for Three Point Bending with Flat Upper Point.

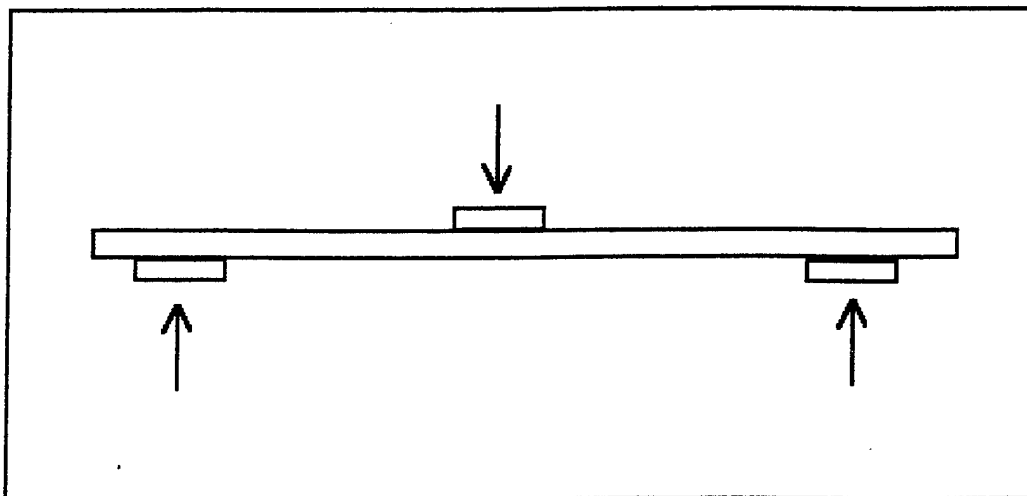


Figure 19. Local Stress Reducer Plates for Three Point Bending.

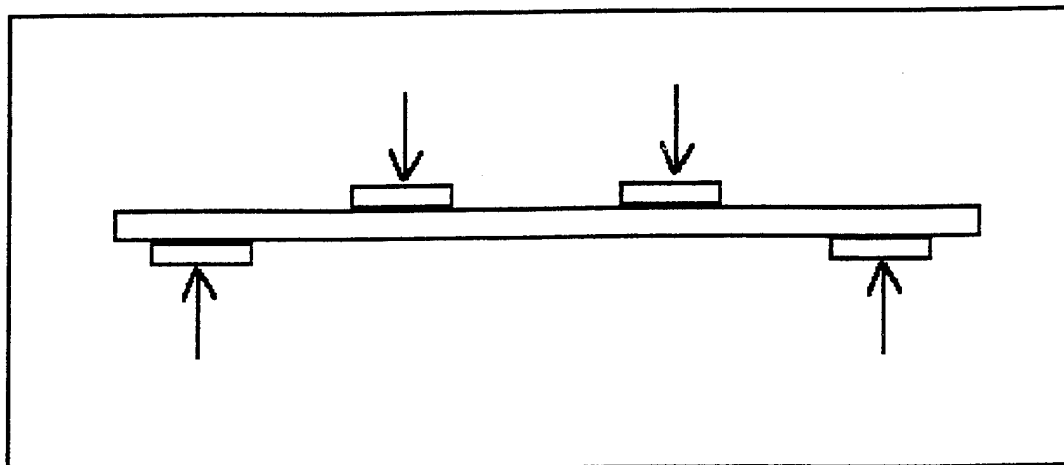


Figure 20. Local Stress Reducer Plates for Four Point Bending.

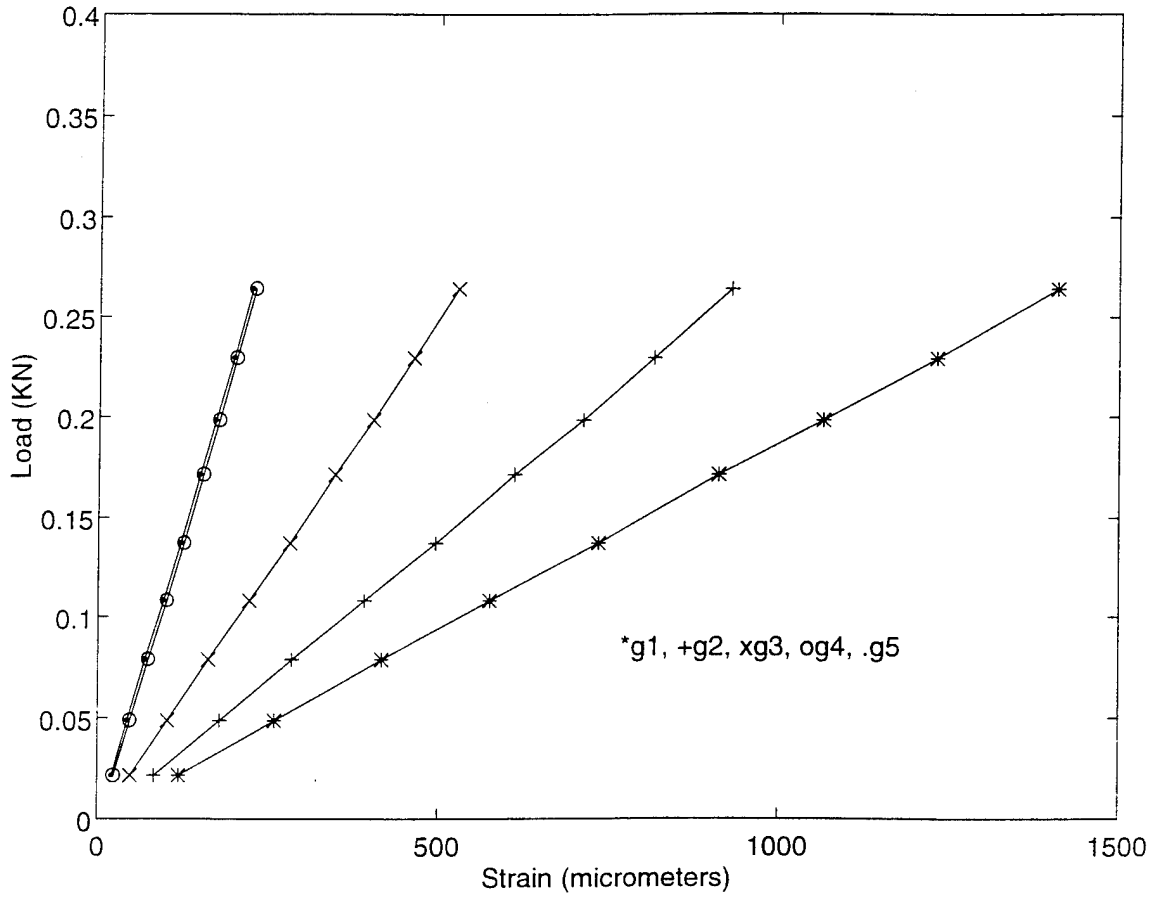


Figure 21. Three Point Bending with 19.05 mm Flat Plates.

| load (KN) | gauge1 | gauge2 | gauge3 | gauge4 | gauge5 |
|-----------|--------|--------|--------|--------|--------|
| 0.02 | 118 | 82 | 48 | 24 | 21 |
| 0.05 | 259 | 178 | 101 | 47 | 43 |
| 0.08 | 417 | 284 | 161 | 73 | 68 |
| 0.11 | 576 | 391 | 221 | 100 | 95 |
| 0.14 | 735 | 496 | 281 | 125 | 120 |
| 0.17 | 912 | 612 | 346 | 153 | 148 |
| 0.20 | 1066 | 712 | 403 | 176 | 171 |
| 0.23 | 1232 | 817 | 463 | 201 | 197 |
| 0.26 | 1410 | 930 | 528 | 229 | 224 |

Table 11. Results for Three Point Bending with 19.05 mm Flat Plates.

| load (KN) | gauge1 | gauge2 | gauge3 | gauge4 | gauge5 |
|-----------|--------|--------|--------|--------|--------|
| 0.03 | 114 | 82 | 48 | 26 | 23 |
| 0.05 | 251 | 178 | 102 | 51 | 48 |
| 0.08 | 397 | 274 | 158 | 76 | 73 |
| 0.11 | 546 | 385 | 218 | 103 | 101 |
| 0.14 | 702 | 495 | 279 | 129 | 129 |
| 0.17 | 862 | 604 | 341 | 155 | 157 |
| 0.20 | 1018 | 709 | 400 | 180 | 183 |
| 0.23 | 118 | 820 | 462 | 206 | 211 |
| 0.27 | 1345 | 930 | 524 | 233 | 239 |

Table 12. Results for Three Point Bending with 25.40 mm Flat Plates.

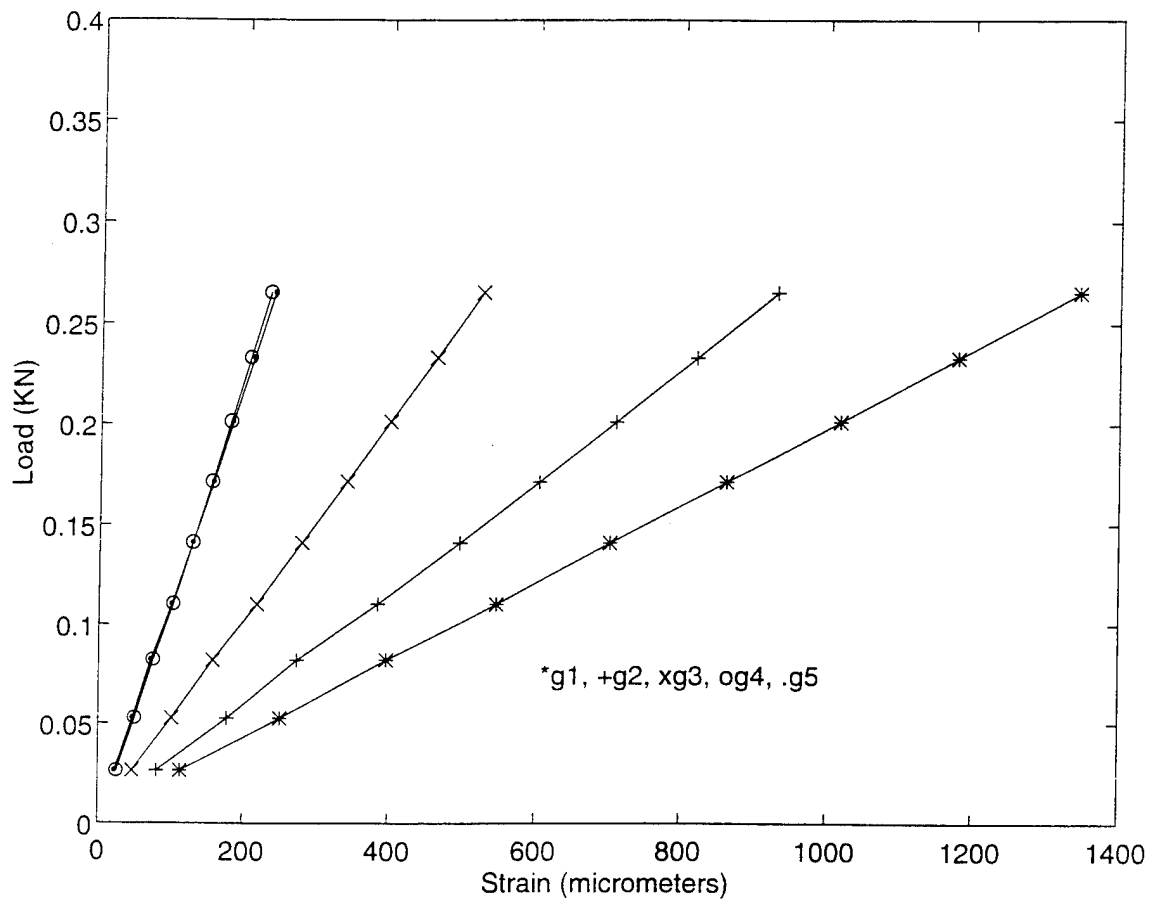


Figure 22. Three Point Bending with 25.40 mm Flat Plates.

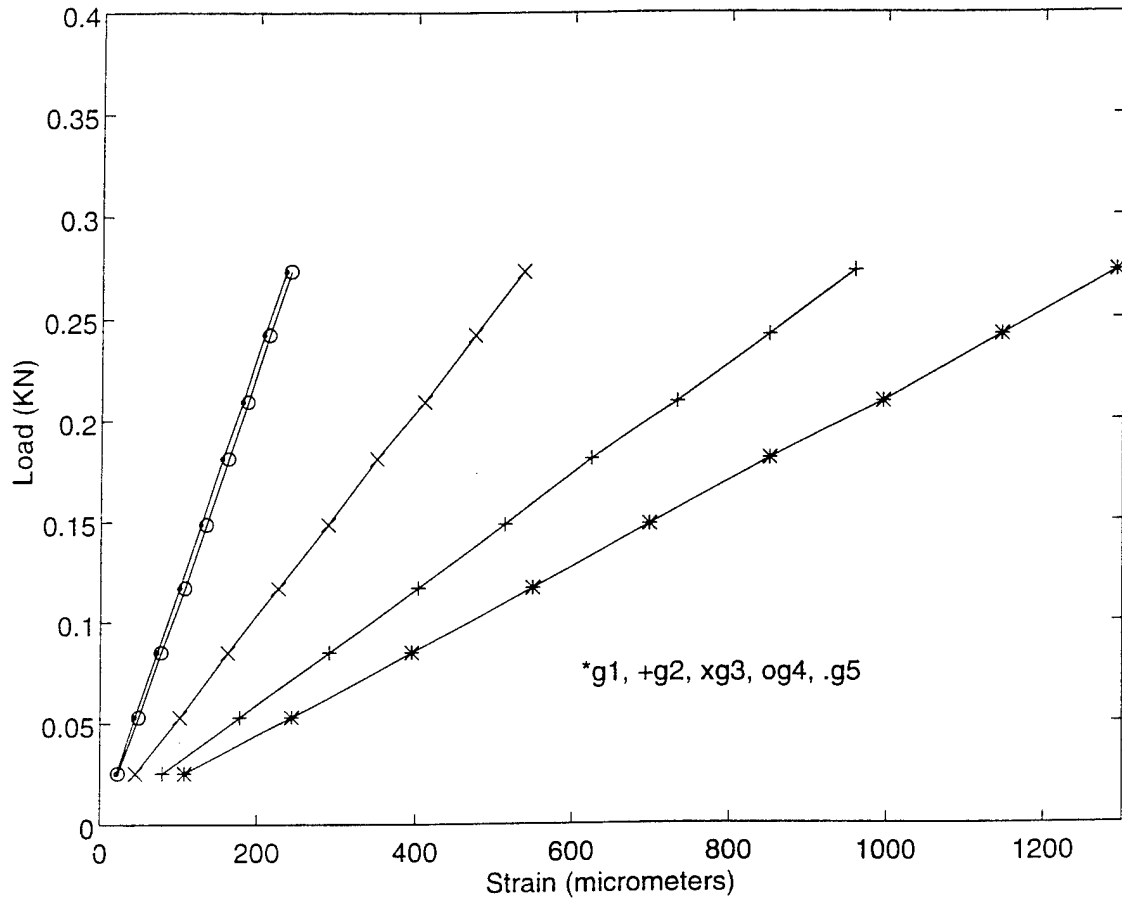


Figure 23. Three Point Bending with 31.75 mm Flat Plates.

| load (KN) | gauge1 | gauge2 | gauge3 | gauge4 | gauge5 |
|-----------|--------|--------|--------|--------|--------|
| 0.03 | 107 | 79 | 45 | 23 | 21 |
| 0.05 | 243 | 178 | 101 | 49 | 44 |
| 0.08 | 395 | 289 | 162 | 77 | 72 |
| 0.12 | 550 | 403 | 226 | 107 | 100 |
| 0.15 | 699 | 514 | 287 | 134 | 127 |
| 0.18 | 850 | 624 | 349 | 162 | 154 |
| 0.21 | 995 | 733 | 410 | 187 | 180 |
| 0.24 | 1147 | 849 | 474 | 214 | 207 |
| 0.27 | 1292 | 959 | 537 | 240 | 234 |

Table 13. Results for Three Point Bending with 31.75 mm Flat Plates.

| load (KN) | gauge1 | gauge2 | gauge3 | gauge4 | gauge5 |
|-----------|--------|--------|--------|--------|--------|
| 0.04 | 73 | 78 | 82 | 33 | 32 |
| 0.08 | 147 | 152 | 164 | 63 | 64 |
| 0.11 | 218 | 224 | 245 | 94 | 95 |
| 0.15 | 289 | 297 | 328 | 125 | 126 |
| 0.18 | 364 | 371 | 414 | 157 | 159 |
| 0.22 | 438 | 448 | 501 | 188 | 191 |
| 0.26 | 521 | 532 | 598 | 222 | 227 |
| 0.29 | 581 | 593 | 669 | 248 | 252 |
| 0.32 | 644 | 656 | 743 | 272 | 280 |

Table 14. Results for Four Point Bending with 25.4 mm Flat Plates.

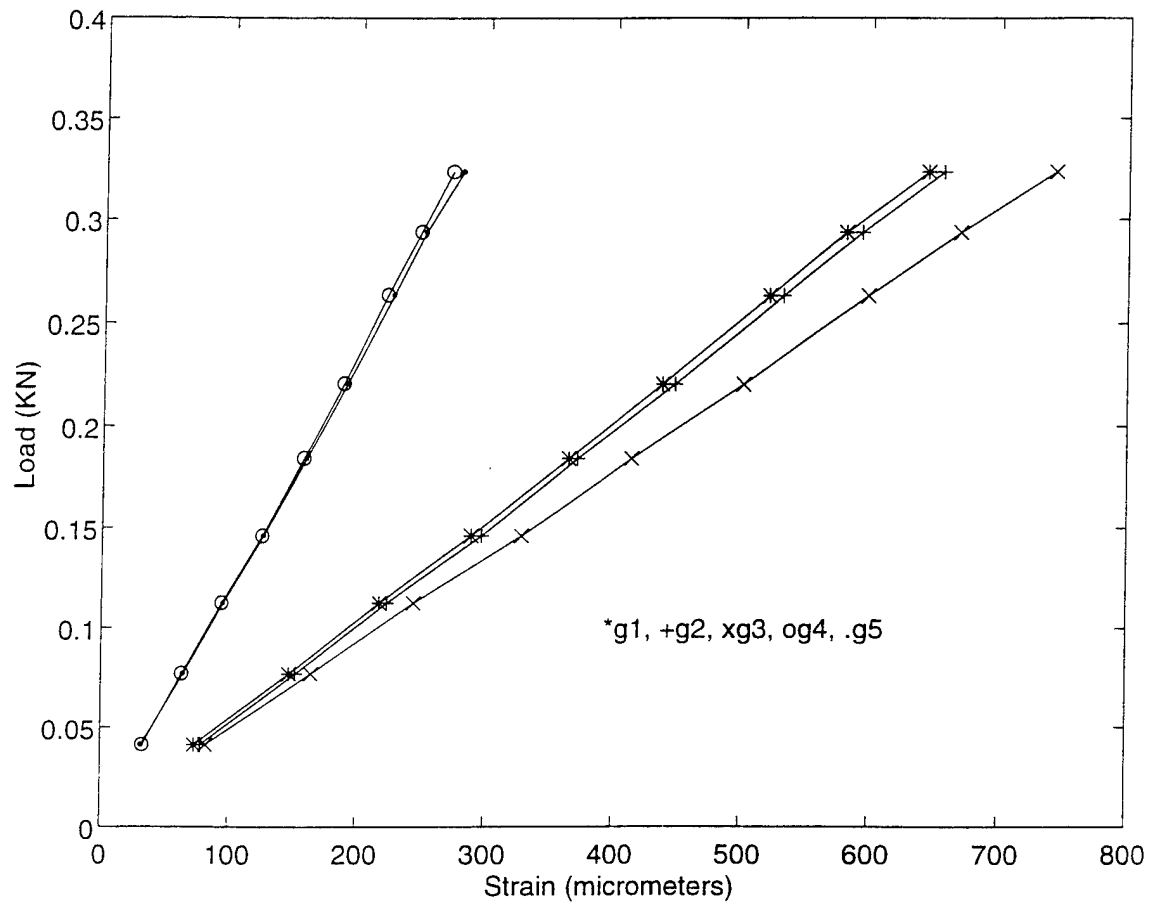


Figure 24. Four Point Bending with 25.4 mm Flat Plates.

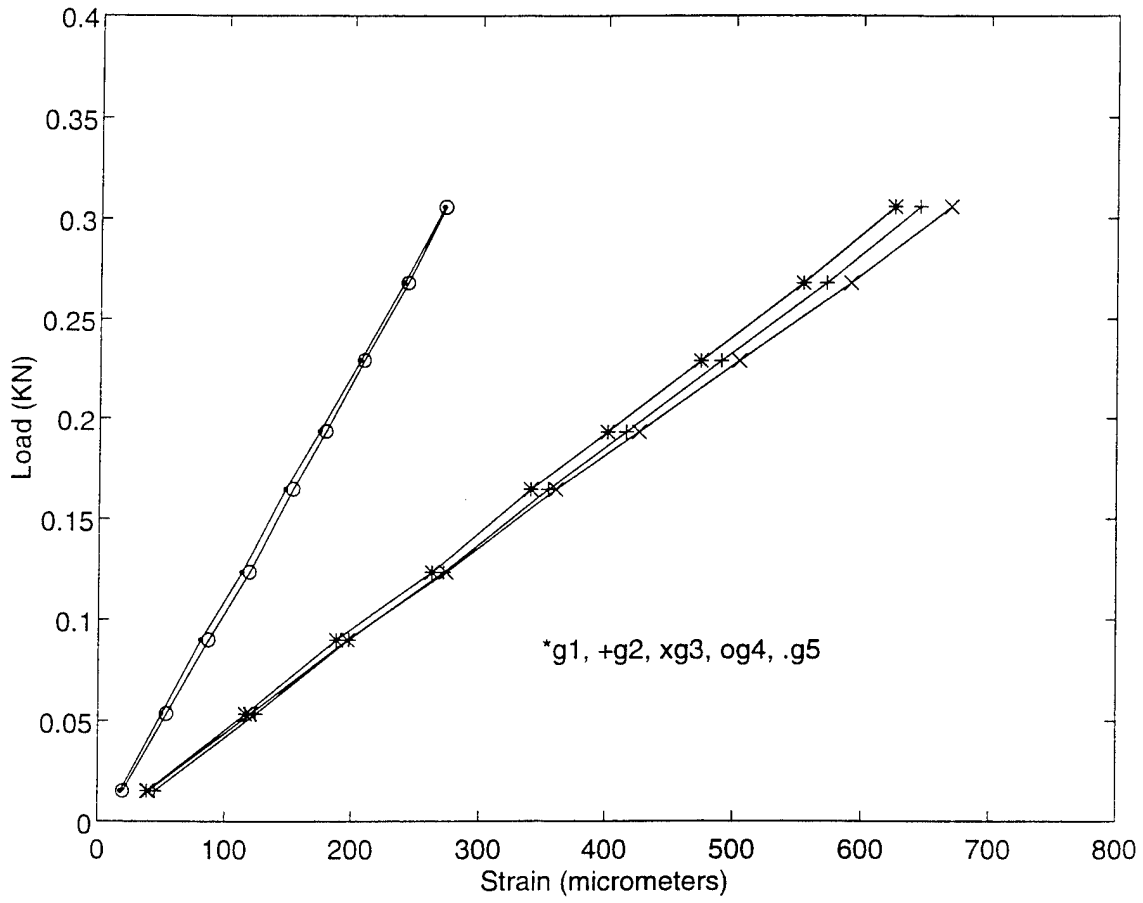


Figure 25. Four Point Bending with 31.75 mm Flat Plates.

| load (KN) | gauge1 | gauge2 | gauge3 | gauge4 | gauge5 |
|-----------|--------|--------|--------|--------|--------|
| 0.02 | 39 | 45 | 40 | 20 | 18 |
| 0.05 | 116 | 124 | 120 | 54 | 50 |
| 0.09 | 188 | 198 | 197 | 87 | 81 |
| 0.12 | 263 | 272 | 274 | 119 | 113 |
| 0.16 | 340 | 354 | 360 | 153 | 147 |
| 0.19 | 401 | 416 | 426 | 179 | 174 |
| 0.23 | 474 | 490 | 504 | 209 | 206 |
| 0.27 | 554 | 572 | 591 | 243 | 240 |
| 0.31 | 625 | 645 | 669 | 272 | 271 |

Table 15. Results for Four Point Bending with 31.75 mm Flat Plates.

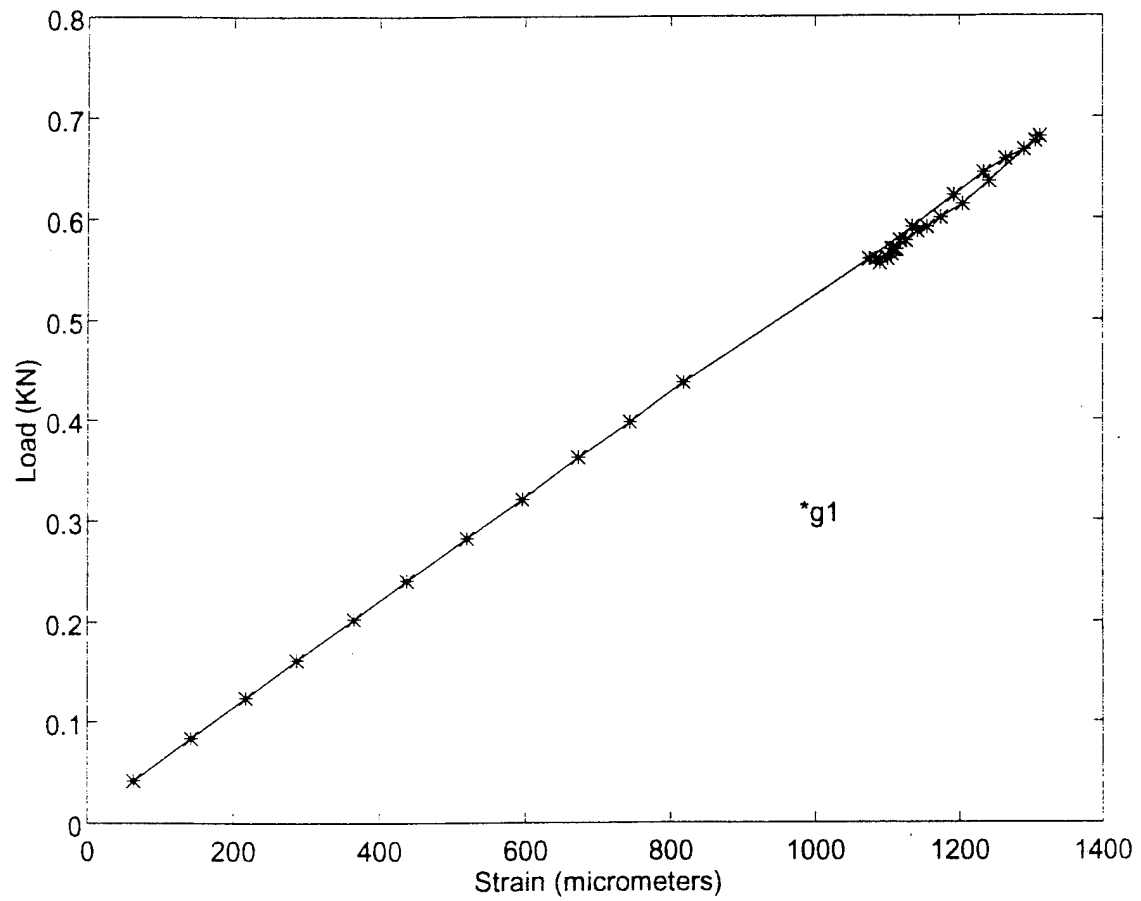


Figure 26. Four Point Bending to Failure with 31.75 mm Flat Plates.

| load (KN) | gauge1 |
|-----------|--------|
| 0.04 | 64 |
| 0.08 | 143 |
| 0.12 | 217 |
| 0.16 | 287 |
| 0.20 | 365 |
| 0.24 | 437 |
| 0.28 | 520 |
| 0.32 | 596 |
| 0.36 | 672 |
| 0.40 | 743 |
| 0.44 | 818 |
| 0.56 | 1075 |
| 0.59 | 1134 |
| 0.62 | 1192 |
| 0.64 | 1233 |
| 0.66 | 1264 |
| 0.67 | 1290 |
| 0.68 | 1306 |
| 0.68 | 1312 |
| 0.67 | 1290 |
| 0.64 | 1241 |
| 0.61 | 1204 |
| 0.60 | 1174 |
| 0.59 | 1155 |
| 0.59 | 1142 |
| 0.58 | 1125 |
| 0.57 | 1112 |

Table 16. Results for Four Point Bending to Failure with 31.75 mm Flat Plates.

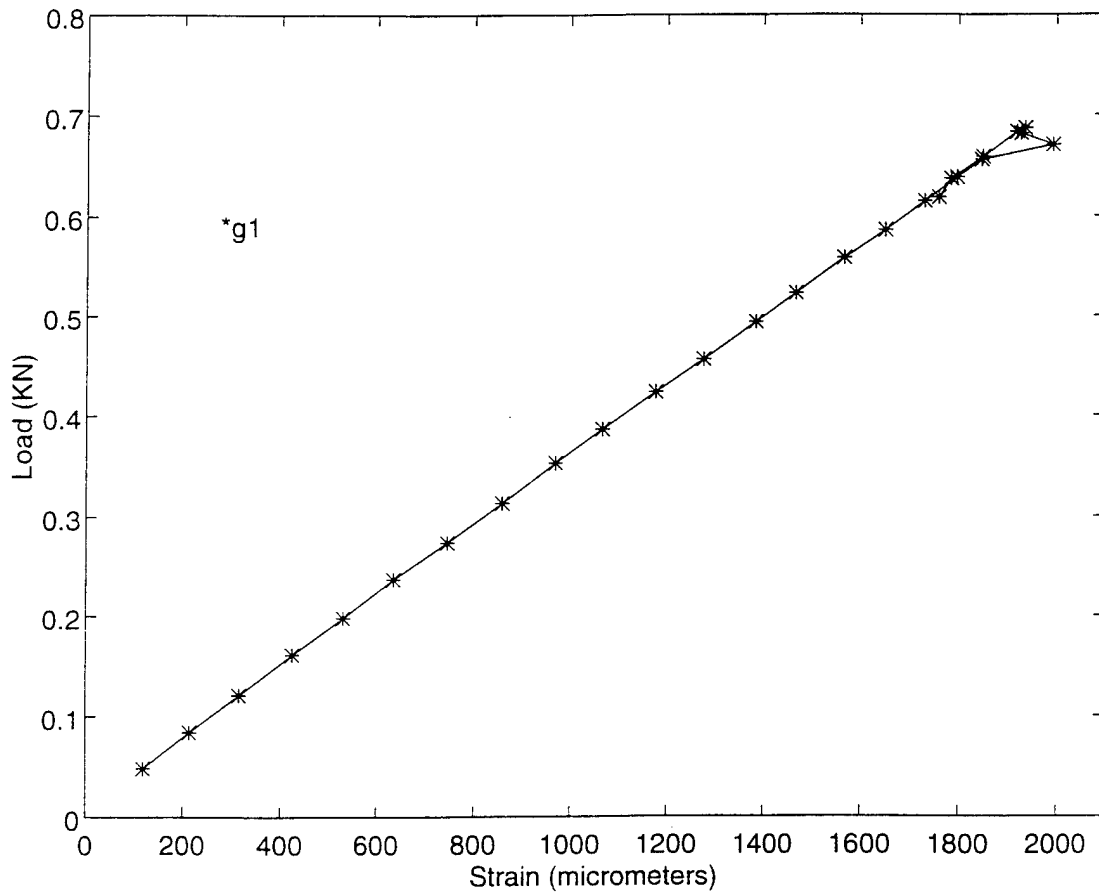


Figure 27. Four Point Bending to Failure with One 12.70 mm Dia. Hole and 31.75 mm Flat Plates.

| load (KN) | gauge1 |
|-----------|--------|
| 0.05 | 119 |
| 0.08 | 214 |
| 0.12 | 316 |
| 0.16 | 425 |
| 0.20 | 530 |
| 0.24 | 636 |
| 0.27 | 746 |
| 0.31 | 859 |
| 0.35 | 970 |
| 0.39 | 1067 |
| 0.42 | 1178 |
| 0.46 | 1276 |
| 0.49 | 1385 |
| 0.52 | 1469 |
| 0.56 | 1568 |
| 0.59 | 1651 |
| 0.62 | 1731 |
| 0.64 | 1797 |
| 0.66 | 1848 |
| 0.67 | 1993 |
| 0.68 | 1927 |
| 0.69 | 1936 |
| 0.68 | 1919 |
| 0.66 | 1850 |
| 0.64 | 1785 |
| 0.62 | 1760 |

Table 17. Results for Four Point Bending to Failure with One 12.70 mm Dia. Hole and 31.75 mm Flat Plates.

V. COMPRESSION TEST

Having determined that the bending test was ineffective in evaluating the effect of stress concentration around a hole in foam cored composite beams, compression testing was pursued using pinned-pinned end conditions as described in Chapter III. The results of the compression testing are not discussed in a chronological order as was the case in chapter IV. First, the tests conducted are listed along with the associated tables and figures. This is followed by a summary of results in table format. A discussion of trends concludes this chapter.

Two tests were conducted on samples with 3 mm (0.12 in.) thick foam. The first sample had a 12.70 mm (0.5 in.) hole at the center. The second sample had a 22.23 mm (0.875 in.) hole at the center. Figures 28 and 29 and Tables 18 and 19 show the results. Strain gauges were located in accordance with Figure 30 for all three tests.

Four tests were conducted on the samples with 6.35 mm (0.25 in.) thick foam. The first two samples had a 12.70 mm (0.5 in.) hole at the center and a 19.05 mm (0.75 in.) hole at the center respectively with strain gauges mounted as shown in Figure 30. See Figures 31 and 32 and Tables 20 and 21 for the test data. The third and fourth samples respectively had 12.70 mm (0.5 in.) and 19.05 mm (0.75 in.) holes at the quarter points. See Figures 33 and 34 and Tables 22 and 23 for the results. The strain gauges were located as shown in Figure 35.

Six tests were conducted on samples with 12.70 mm (0.5 in.) thick foam. The first three samples had a 6.35 mm (0.25 in.) hole at the center, a 12.70 mm (0.5 in.) hole at the center, and a 19.05 mm (0.75 in.) hole at the center respectively. See Figures 36, 37, and 38 and Tables 24, 25, and 26 for the results. Strain gauge locations are shown in Figure 30. The fourth sample had a 6.35 mm (0.25 in.) hole at one quarter point with strain gauge locations as in Figure 39. See Figure 40 and Table 27. The fifth sample had a 6.35 mm (0.25 in.) hole at both quarter points with strain gauge locations as shown in Figure 35. See Figure 41 and Table 28. The sixth sample had a 6.35 mm (0.25 in.) hole at each quarter point and at the center, with strain gauge locations as shown in Figure 42. The results are shown in Figure 43 and Table 29.

Referring to Table 30, the compression test results will now be discussed. Prior to failure, each compressed sample took a characteristic shape as shown in Figure 44. The strain diagrams indicate a trend described by initial column compression followed by column bending.

The sample with a core thickness of 3 mm (0.12 in.) without a circular hole failed at a load of 3.47 KN at the quarter point, [Ref. 2]. The same size sample when subjected to a 12.70 mm (0.5 in.) circular hole at the center failed at 2.74 KN at the quarter point for a load reduction of 21 percent. When subjected to a 22.23 mm (0.875 in.) hole, the sample failed at 2.61 KN at the center for a reduction of 24.8 percent. When failure occurred at the quarter point away from the hole, foam core shear failure and delamination were evident. The carbon skin was left intact. When failure occurred at the hole, failure by bending was evident as the

carbon skin also failed.

The sample with a 6.35 mm (0.25 in.) thick core without a hole failed at 6.31 KN at the quarter point, [Ref. 2]. With a 12.70 mm (0.5 in.) centered hole, it failed at 6.13 KN at the quarter point for a 3.0 percent reduction. When the hole was increased to 19.05 mm (0.75 in.) the sample failed at 5.24 KN at the center for a reduction of 17.1 percent. The tests conducted on the samples with a core thickness of 6.35 mm (0.25 in.) followed the same trend as the tests conducted on the sample with a core thickness of 3 mm (0.12 in.). Failure by bending occurred when the hole diameter was 19.05 mm (0.75 in.), vice 22.23 mm (.875 in.) for the sample with the 3 mm (0.12 in.) thick core.

The sample with a 12.70 mm (0.5 in.) thick core without a hole failed at 14.23 KN at the quarter point, [Ref. 2]. A 6.35 mm (0.25 in.) hole reduced the failure load by 13.5 percent. The transition to failing at the center occurred when subjected to a 12.70 mm (0.5 in.) hole at the center. This resulted in a failure load reduction of 28.1 percent. The tests show that a foam cored sandwich composite beam subjected to stress concentration at a hole and loaded in compression with pinned-pinned end conditions fails at a reduced load. The beam fails at the quarter point by foam core shear failure until the hole reaches a critical diameter at which point the beam fails by bending failure at the hole. The critical diameter decreases as the foam core thickness increases.

Further tests were conducted locating the circular hole at the quarter point of the beam. Using a sample with a core thickness of 6.35 mm (0.25 in.), a test was conducted with a 12.70 mm (0.5 in.) hole at each quarter point. The beam failed at 5.74 KN at the quarter

point for a reduction in failure load of 9.2 percent when compared to the no-hole sample. When this sample was subjected to a 19.05 mm (0.75 in.) hole at each quarter point, the failure occurred at 4.99 KN at the quarter point for a reduction of 21.0 percent. In general, the failure load decreased when the location of a given diameter hole was moved from the center to the quarter point, with failure occurring at the quarter point.

Similar tests were conducted on the samples with a foam thickness of 12.70 mm (0.5 in.). A sample with a 6.35 mm (0.25 in.) hole at one quarter point failed at 13.41 KN at the quarter point. A sample with a 6.35 mm (0.25 in.) hole at each quarter point failed at 13.21 KN at the quarter point. A sample with 6.35 mm (0.25 in.) holes at the center and at both quarter points failed at 13.15 KN at the quarter point. These failure loads are approximately 7 percent lower than the no-hole failure load. Unlike the samples with 6.35 mm (0.25 in.) thick cores, the samples with 12.70 mm (0.5 in.) thick cores failed at slightly higher loads when the hole was moved from the center to the quarter point. The experimental failure load for the sample with a 6.35 mm (0.25 in.) hole at the center is inconsistent compared to other test results. This may be due to a variation in specimen geometry or an experimental error. It is recommended that this test be repeated for accuracy.

Having established the trends within each sample thickness category, the discussion will now turn to an analysis across sample thickness groups, holding circular hole locations constant.

The 6.35 mm (0.25 in.) thick foam is 112 percent thicker than the 3 mm (0.12 in.) foam while the 12.70 mm (0.5 in.) foam is 100 percent thicker than the 6.35 mm (0.25 in.)

foam. With no hole, the failure load increases by 82 percent when increasing foam thickness from 3mm (0.12 in.) to 6.35mm (0.25 in.) and by 125 percent when increasing from 6.35 mm (0.25 in.) to 12.70 mm (0.5 in.).

When subjected to a 12.70 mm (0.5 in.) hole at the center, the failure load increased by 124 percent when increasing foam thickness from 3 mm (0.12 in.) to 6.35 mm (0.5 in.). A comparison cannot be made to the sample with 12.70 mm (0.5 in.) thick foam as the sample failed at the center vice the quarter point.

When subjected to a 19.05 mm (0.75 in.) hole at the center, the failure load increased by 96 percent when increasing from a foam thickness of 6.35 mm (0.25 in.) to 12.70 mm (0.5 in.).

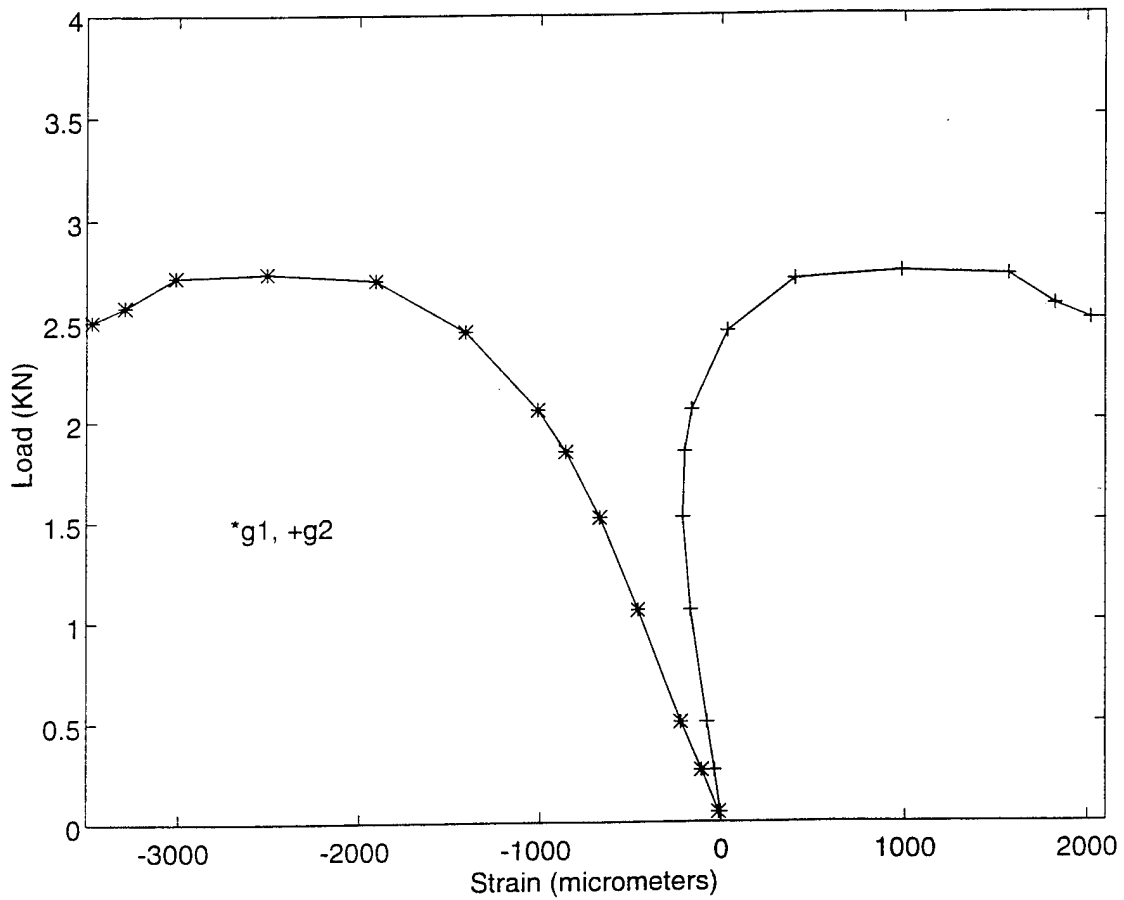


Figure 28. Compression Test, 3 mm Thick, One 12.70 mm Dia. Hole at the Center.

| load (KN) | gauge1 | gauge2 |
|-----------|--------|--------|
| 0.05 | -14 | -5 |
| 0.26 | -106 | -36 |
| 0.50 | -219 | -78 |
| 1.05 | -460 | -168 |
| 1.52 | -675 | -210 |
| 1.85 | -861 | -200 |
| 2.06 | -1013 | -162 |
| 2.45 | -1412 | 33 |
| 2.70 | -1909 | 402 |
| 2.74 | -2511 | 984 |
| 2.72 | -3016 | 1565 |
| 2.57 | -3290 | 1823 |
| 2.50 | -3470 | 2020 |

Table 18. Compression Test, 3 mm Thick, One 12.70 mm Dia. Hole at the Center.

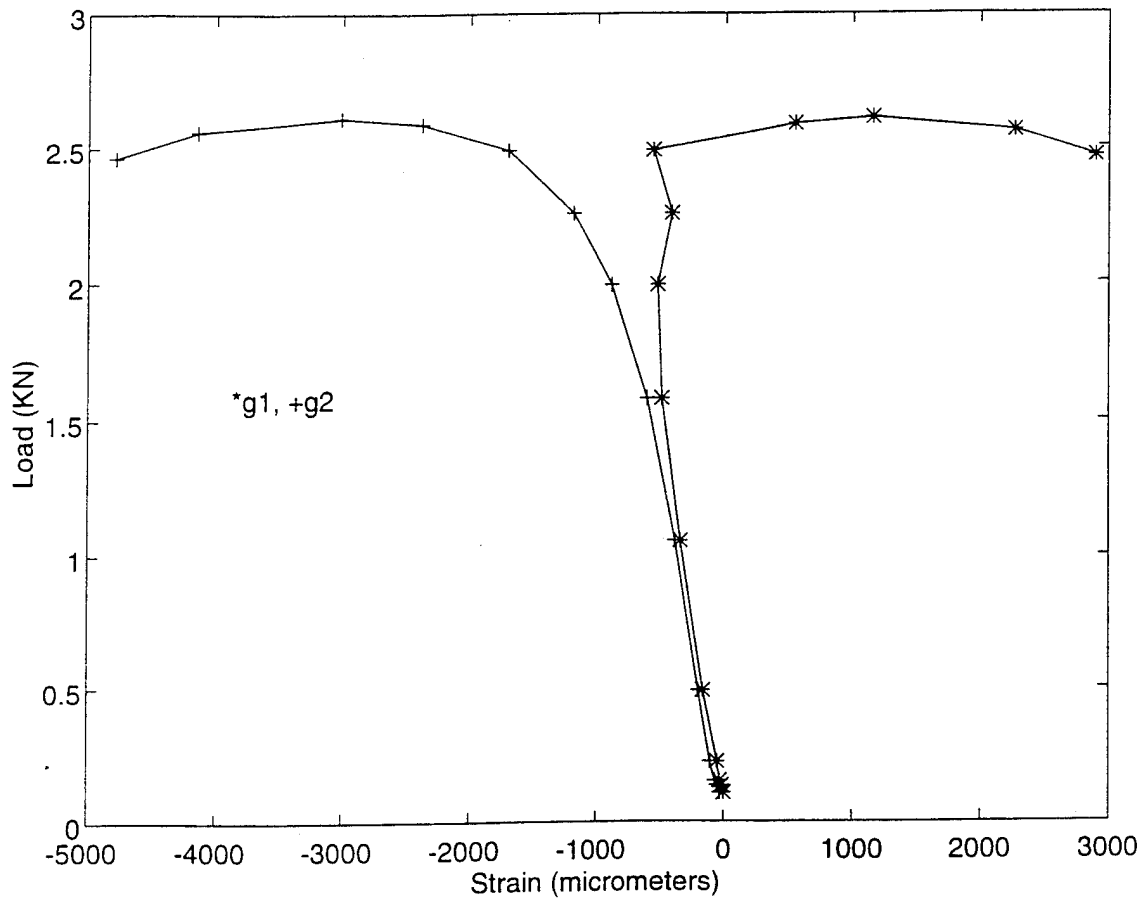


Figure 29. Compression Test, 3 mm Thick, One 22.23 mm Dia. Hole at the Center.

| load (KN) | gauge1 | gauge2 |
|-----------|--------|--------|
| 0.11 | 0 | -30 |
| 0.13 | -15 | -52 |
| 0.15 | -30 | -70 |
| 0.22 | -50 | -105 |
| 0.49 | -163 | -199 |
| 1.05 | -340 | -382 |
| 1.58 | -484 | -597 |
| 1.99 | -520 | -880 |
| 2.26 | -414 | -1188 |
| 2.49 | -56 | -1704 |
| 2.59 | 545 | -2382 |
| 2.61 | 1155 | -3014 |
| 2.56 | 2268 | -4146 |
| 2.46 | 2896 | -4786 |

Table 19. Compression Test, 3 mm Thick, One 22.23 mm Dia. Hole at the Center..

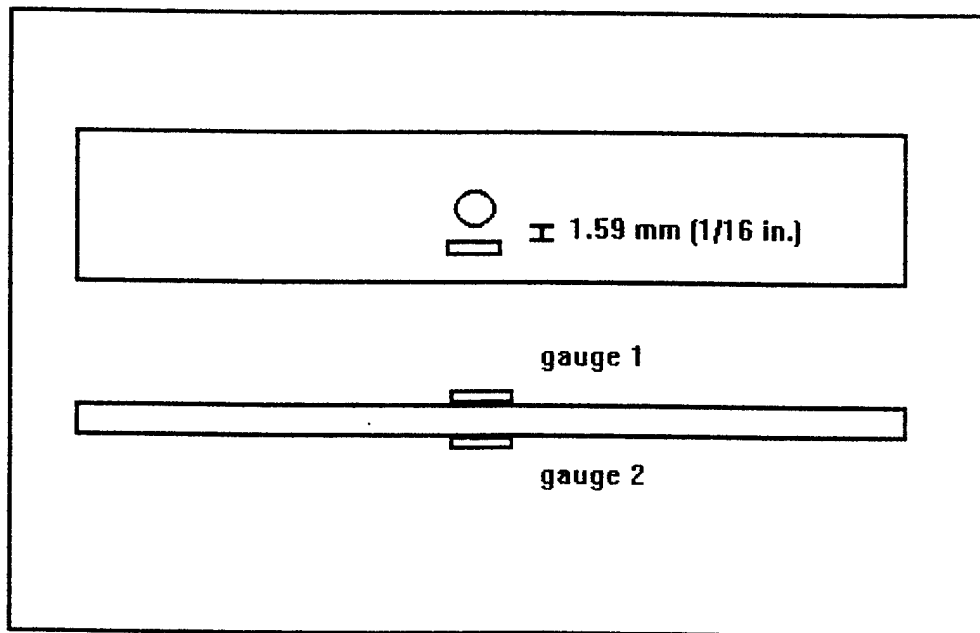


Figure 30. Strain Gauge Locations for 3 mm Thick Compression Tests.

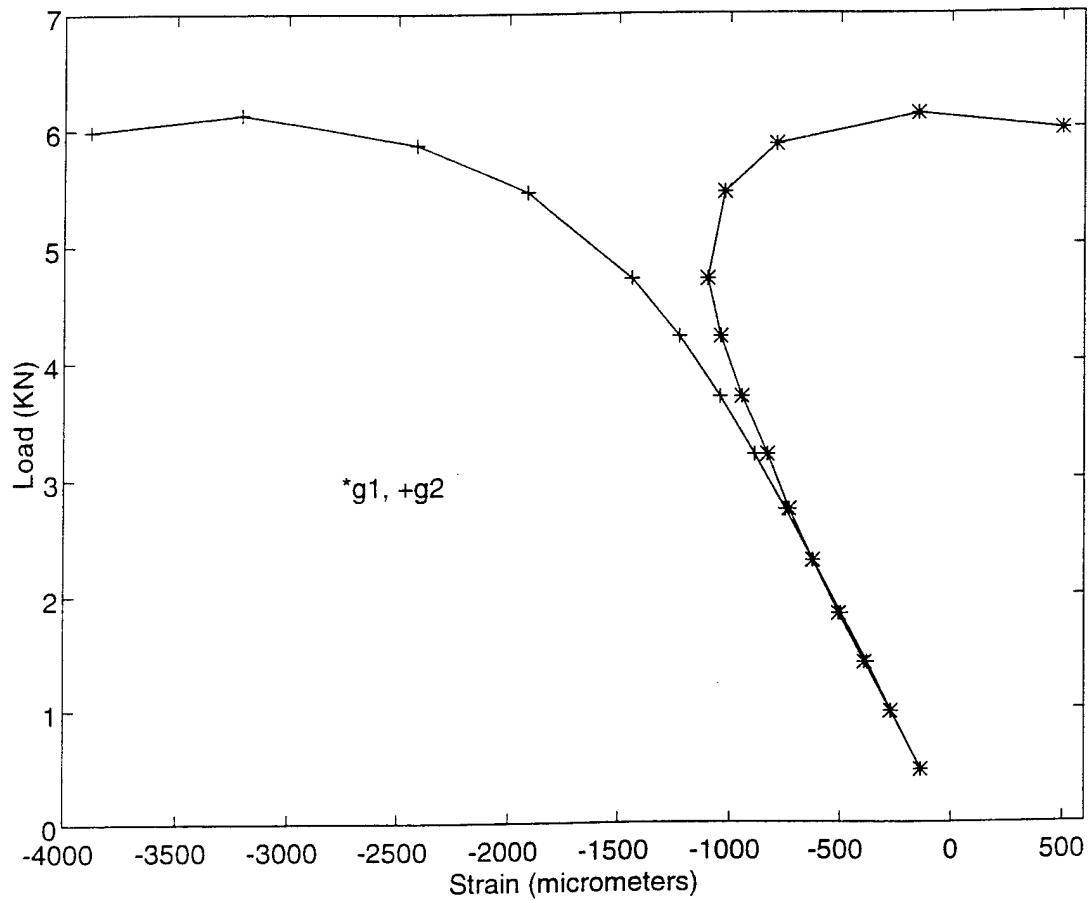


Figure 31. Compression Test, 6.35 mm Thick, One 12.70 mm Dia. Hole at the Center.

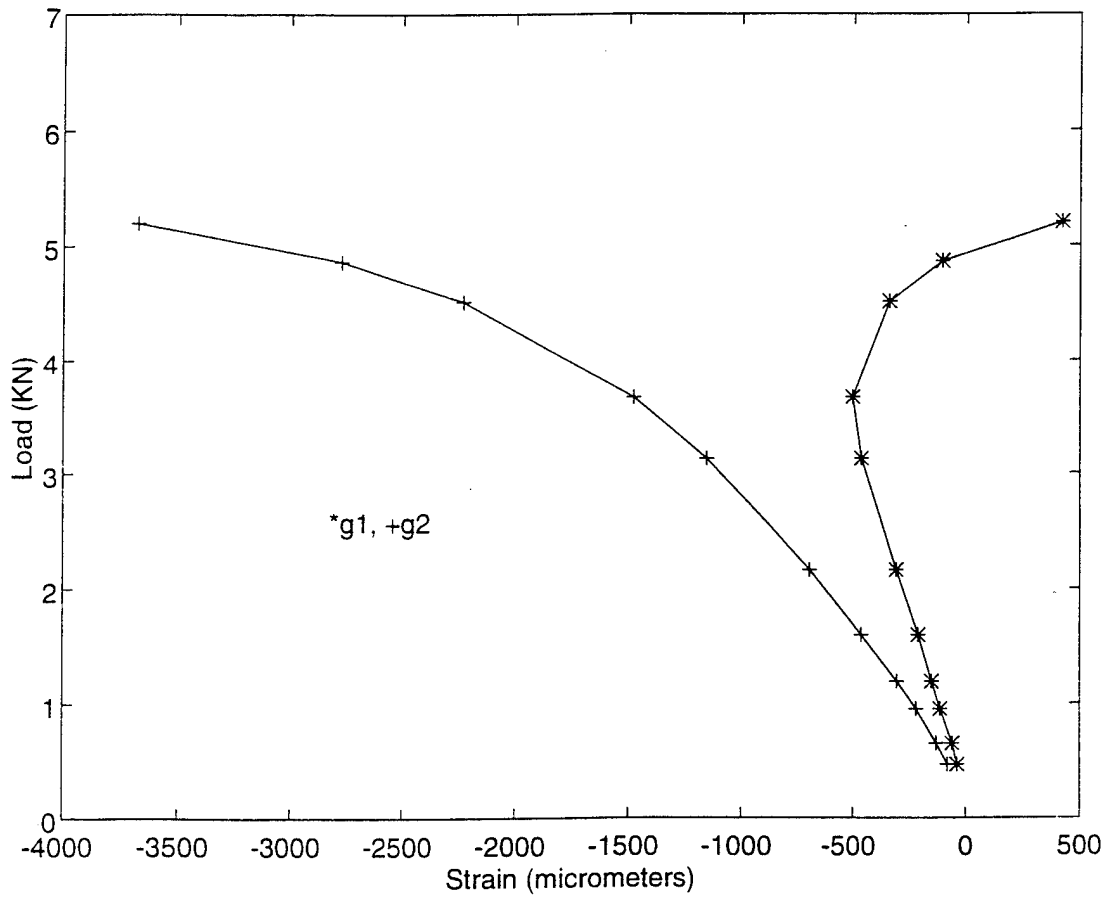


Figure 32. Compression Test, 6.35 mm Thick, One 19.05 mm Dia. Hole at the Center.

| load (KN) | gauge1 | gauge2 |
|-----------|--------|--------|
| 0.46 | -135 | -135 |
| 0.98 | -272 | -270 |
| 1.41 | -391 | -380 |
| 1.84 | -506 | -497 |
| 2.30 | -624 | -622 |
| 2.74 | -732 | -749 |
| 3.21 | -830 | -890 |
| 3.70 | -947 | -1045 |
| 4.22 | -1043 | -1228 |
| 4.72 | -1102 | -1440 |
| 5.46 | -1026 | -1914 |
| 5.87 | -792 | -2417 |
| 6.13 | -150 | -3209 |
| 5.99 | 500 | -3884 |

Table 20. Compression Test, 6.35 mm Thick, One 12.70 mm Dia. Hole at the Center.

| load (KN) | gauge1 | gauge2 |
|-----------|--------|--------|
| 0.47 | -37 | -80 |
| 0.65 | -59 | -130 |
| 0.96 | -112 | -218 |
| 1.19 | -150 | -303 |
| 1.60 | -210 | -463 |
| 2.16 | -307 | -694 |
| 3.13 | -463 | -1153 |
| 3.68 | -503 | -1477 |
| 4.52 | -339 | -2229 |
| 4.87 | -105 | -2775 |
| 5.20 | 420 | -3672 |
| 5.24 | | -4689 |

Table 21. Compression Test, 6.35 mm Thick, One 19.05 mm Dia. Hole at the Center.

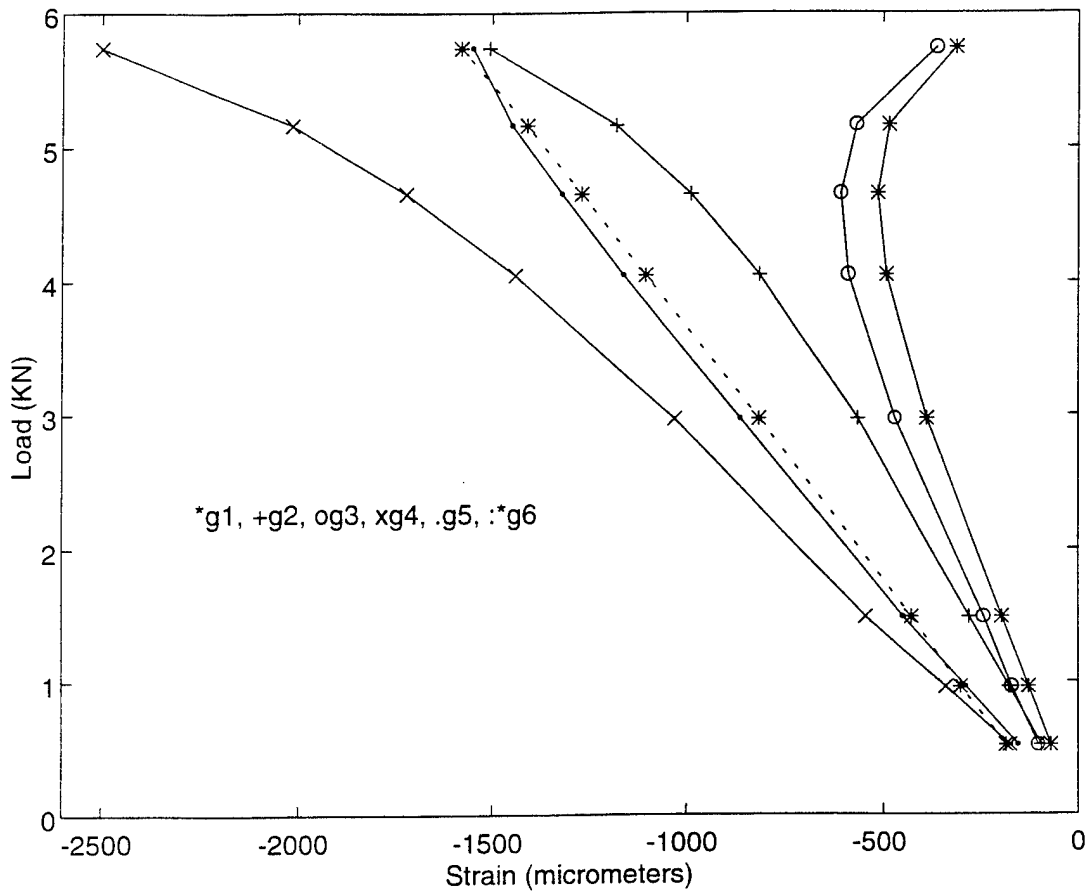


Figure 33. Compression Test, 6.35 mm Thick, One 12.70 mm Dia. Hole at Each Quarter Point.

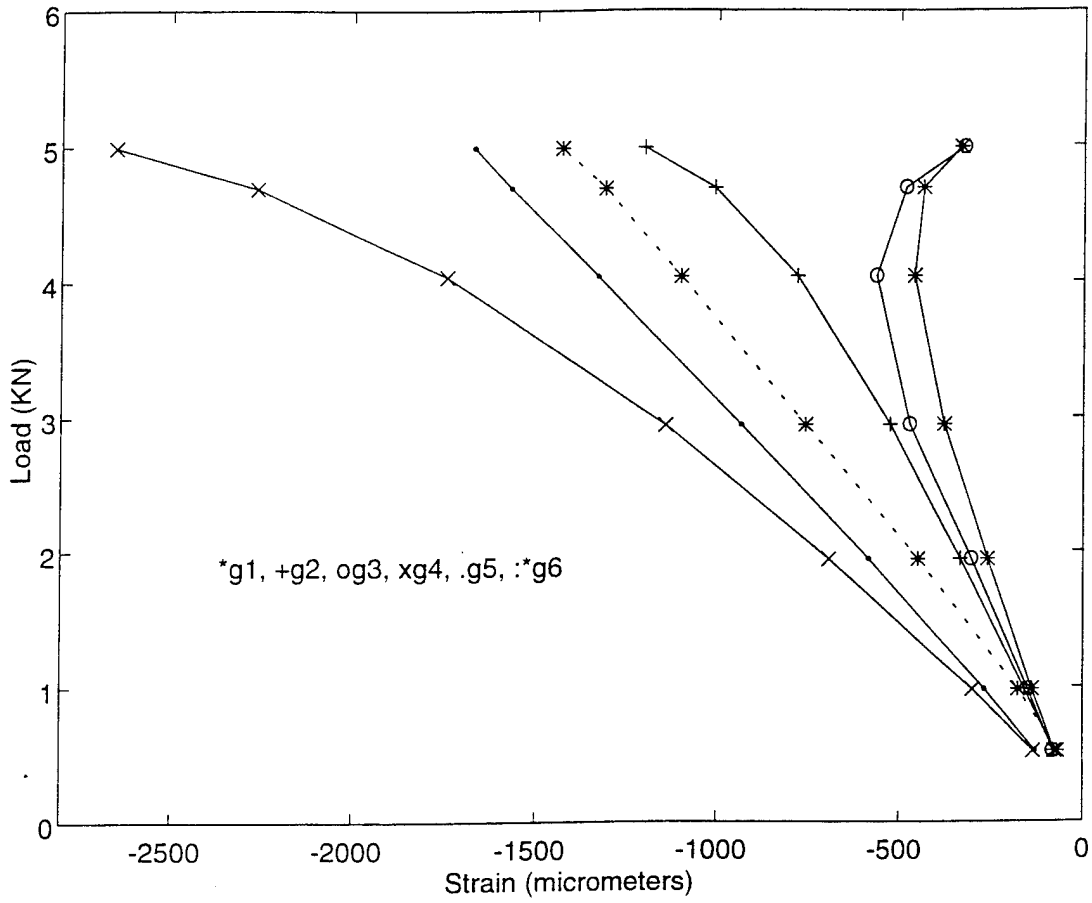


Figure 34. Compression Test, 6.35 mm Thick, One 19.05 mm Dia. Hole at Each Quarter Point.

| load (KN) | gauge1 | gauge2 | gauge3 | gauge4 | gauge5 | gauge6 |
|-----------|--------|--------|--------|--------|--------|--------|
| 0.52 | -70 | -95 | -103 | -175 | -154 | -185 |
| 0.96 | -127 | -177 | -171 | -340 | -288 | -302 |
| 1.49 | -196 | -280 | -243 | -546 | -449 | -427 |
| 2.97 | -388 | -566 | -470 | -1030 | -863 | -815 |
| 4.05 | -492 | -814 | -590 | -1439 | -1159 | -1103 |
| 4.65 | -514 | -989 | -609 | -1720 | -1320 | -1269 |
| 5.16 | -485 | -1179 | -569 | -2015 | -1448 | -1410 |
| 5.74 | -313 | -1507 | -362 | -2500 | -1549 | -1579 |

Table 22. Compression Test, 6.35 mm Thick, One 12.70 mm Dia. Hole at Each Quarter Point.

| load (KN) | gauge1 | gauge2 | gauge3 | gauge4 | gauge5 | gauge6 |
|-----------|--------|--------|--------|--------|--------|--------|
| 0.52 | -76 | -78 | -80 | -133 | -128 | -68 |
| 0.98 | -137 | -157 | -148 | -299 | -266 | -175 |
| 1.94 | -258 | -333 | -303 | -694 | -585 | -448 |
| 2.95 | -377 | -526 | -471 | -1140 | -933 | -758 |
| 4.04 | -460 | -782 | -565 | -1744 | -1328 | -1100 |
| 4.69 | -435 | -1008 | -483 | -2266 | -1569 | -1309 |
| 4.99 | -333 | -1199 | -324 | -2652 | -1670 | -1430 |

Table 23. Compression Test, 6.35 mm Thick, One 19.05 mm Dia. Hole at Each Quarter Point.

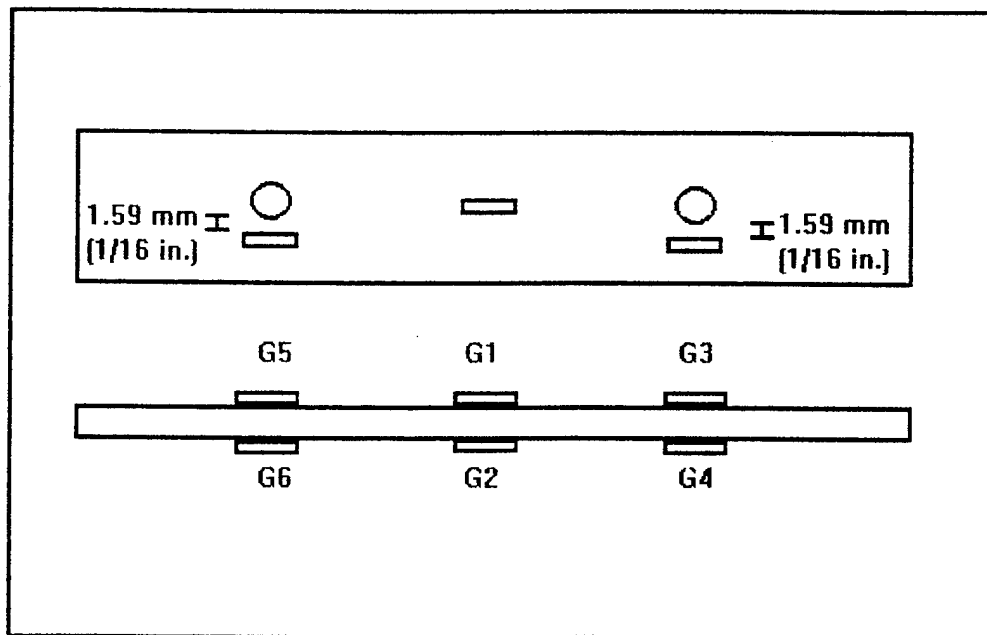


Figure 35. Strain Gauge Locations for Samples with Holes at Quarter Points.

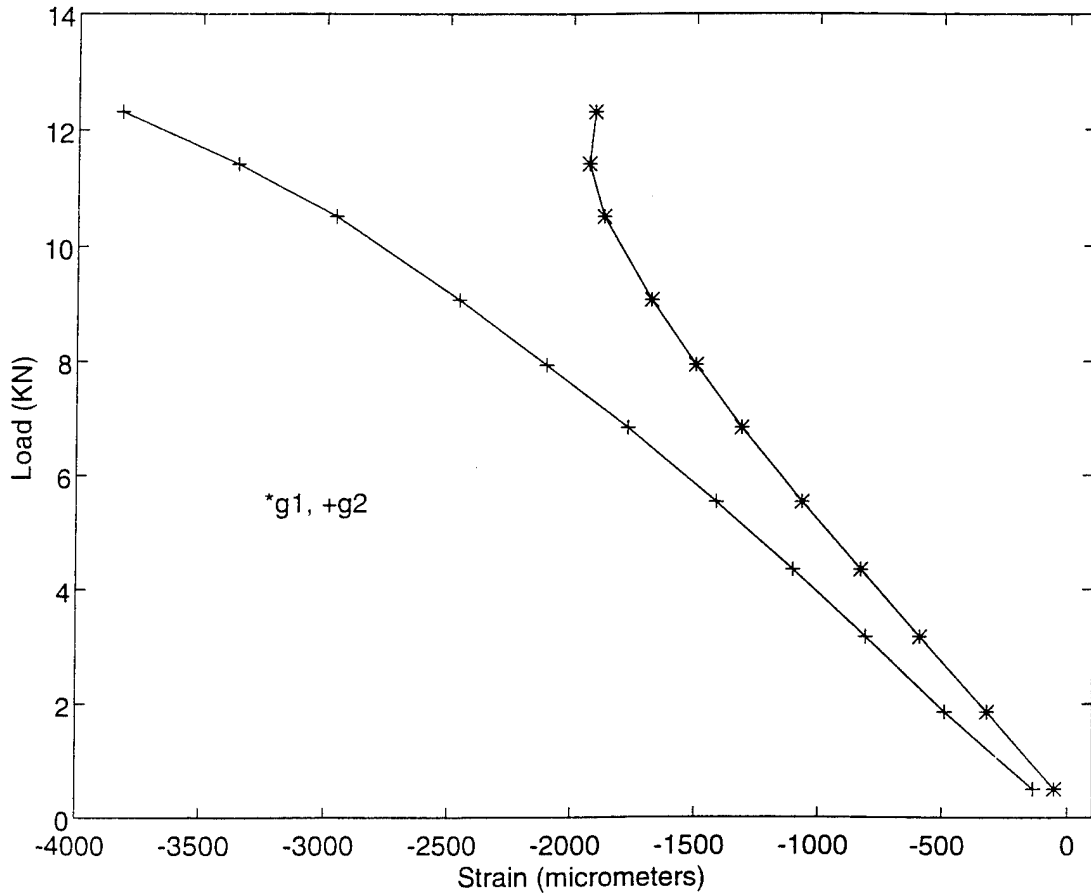


Figure 36. Compression Test, 12.70 mm Thick, 6.35 mm Dia. Hole at the Center.

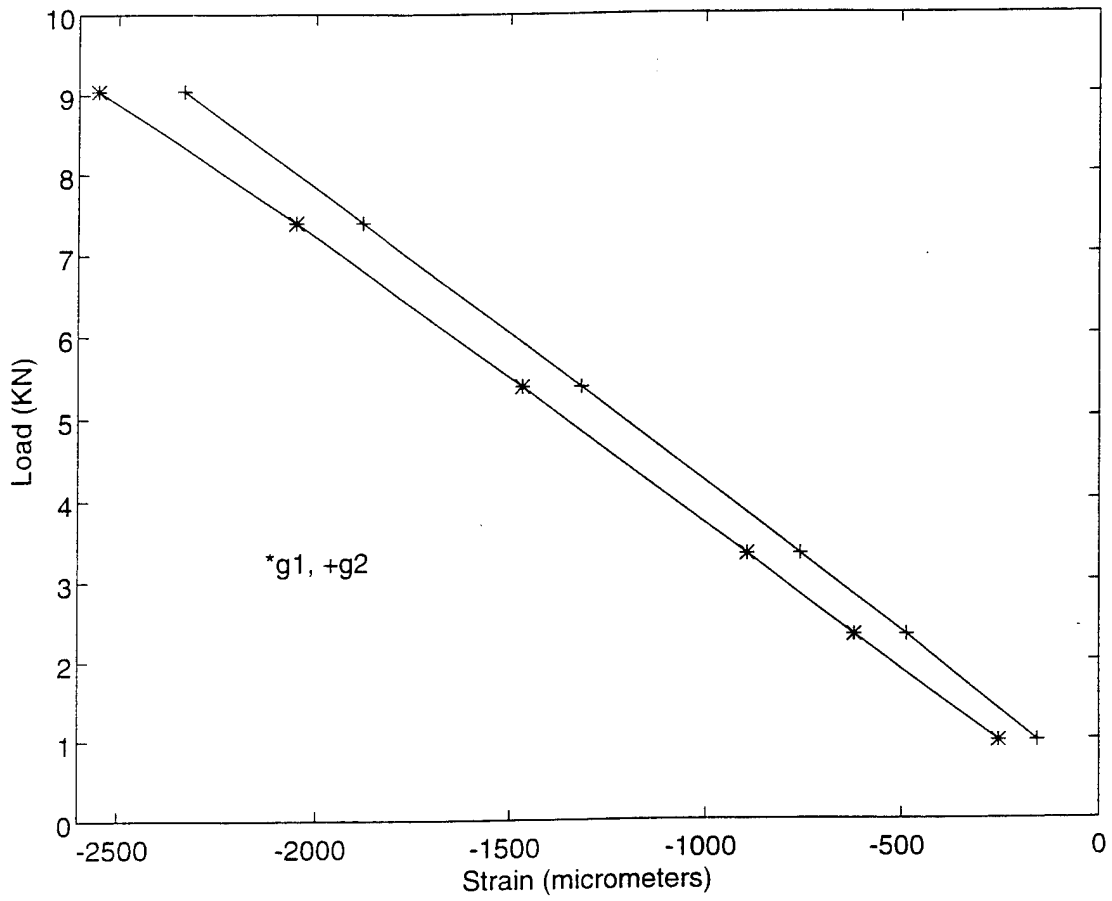


Figure 37. Compression Test, 12.70 mm Thick, 12.70 mm Dia. Hole at the Center.

| load (KN) | gauge1 | gauge2 |
|-----------|--------|--------|
| 0.50 | -52 | -135 |
| 1.86 | -319 | -489 |
| 3.17 | -590 | -809 |
| 4.35 | -829 | -1102 |
| 5.54 | -1066 | -1413 |
| 6.84 | -1311 | -1770 |
| 7.94 | -1497 | -2100 |
| 9.07 | -1677 | -2456 |
| 10.52 | -1822 | -2958 |
| 11.42 | -1934 | -3353 |
| 12.31 | -1911 | -3820 |

Table 24. Compression Test, 12.70 mm Thick, 6.35 mm Dia. Hole at the Center.

| load (KN) | gauge1 | gauge2 |
|-----------|--------|--------|
| 0.98 | -254 | -157 |
| 2.33 | -620 | -487 |
| 3.34 | -895 | -757 |
| 5.39 | -1468 | -1317 |
| 7.40 | -2049 | -1878 |
| 9.04 | -2550 | -2335 |
| 10.23 | | |

Table 25. Compression Test, 12.70 mm Thick, 12.70 mm Dia. Hole at the Center.

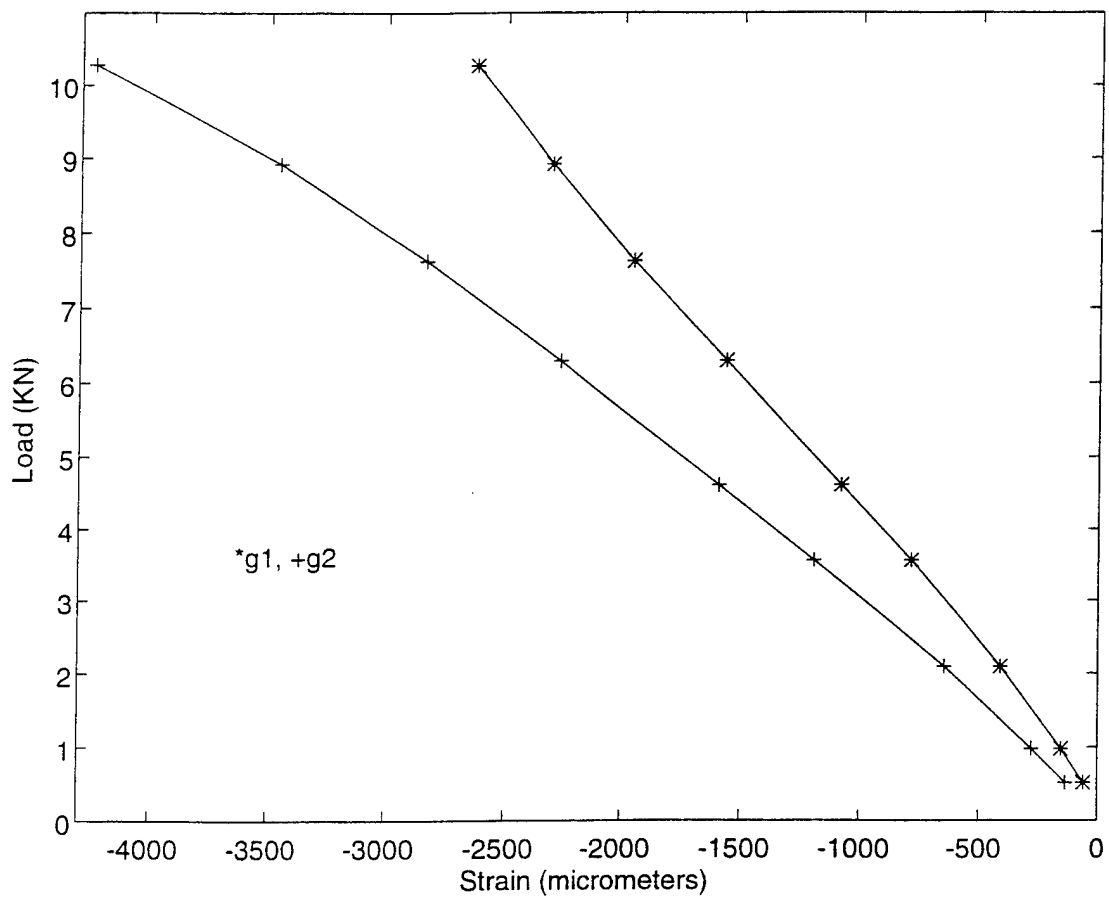


Figure 38. Compression Test, 12.70 mm Thick, 19.05 mm Dia. Hole at the Center.

| load (KN) | gauge1 | gauge2 |
|-----------|--------|--------|
| 0.51 | -58 | -134 |
| 0.98 | -152 | -277 |
| 2.10 | -410 | -647 |
| 3.56 | -786 | -1190 |
| 4.62 | -1078 | -1593 |
| 6.31 | -1564 | -2264 |
| 7.63 | -1953 | -2834 |
| 8.93 | -2303 | -3455 |
| 10.28 | -2629 | -4244 |

Table 26. Compression Test, 12.70 mm Thick, 19.05 mm Dia. Hole at the Center.

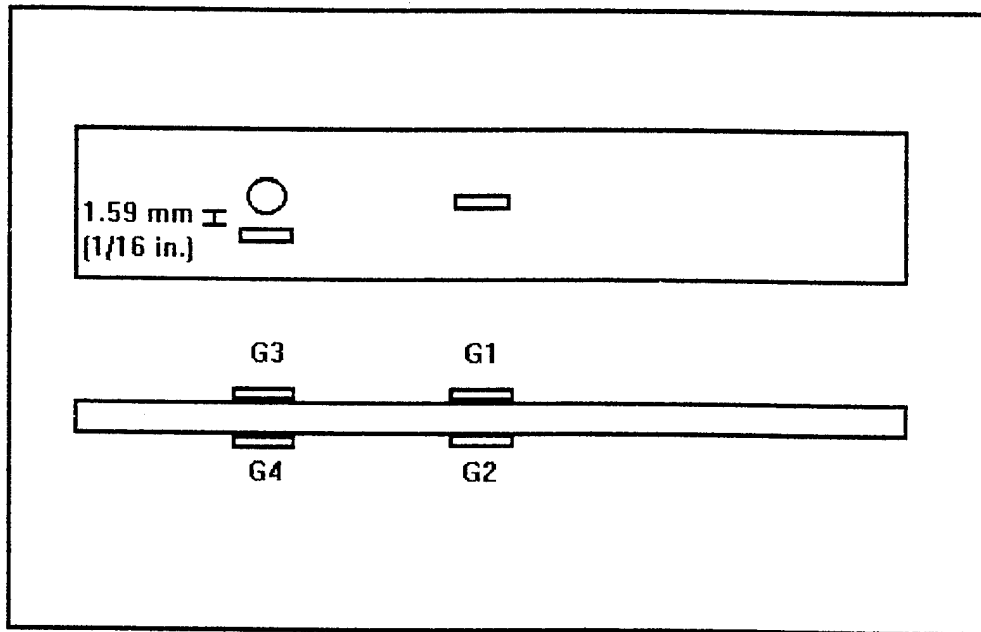


Figure 39. Strain Gauge Location for Samples with One Hole at One Quarter Point.

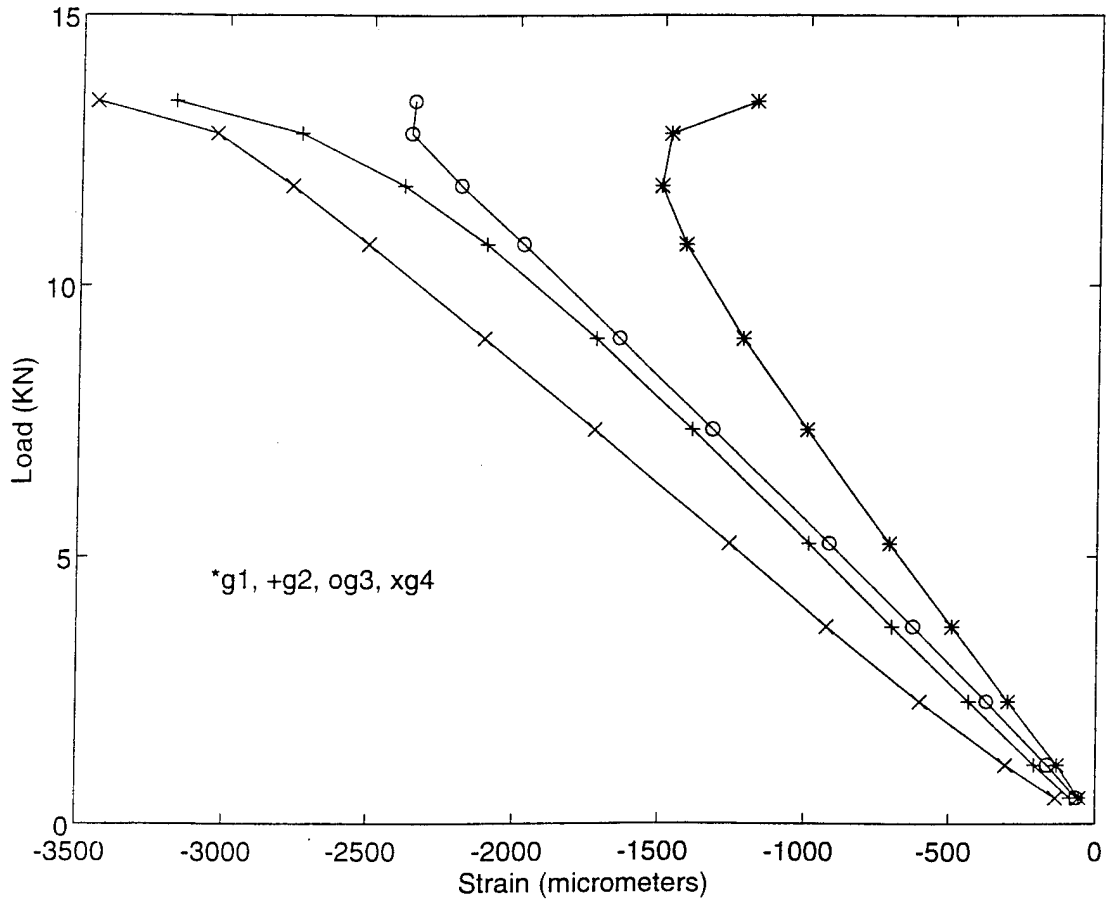


Figure 40. Compression Test, 12.70 mm Thick, One 6.35 mm Dia. Hole at One Quarter Point.

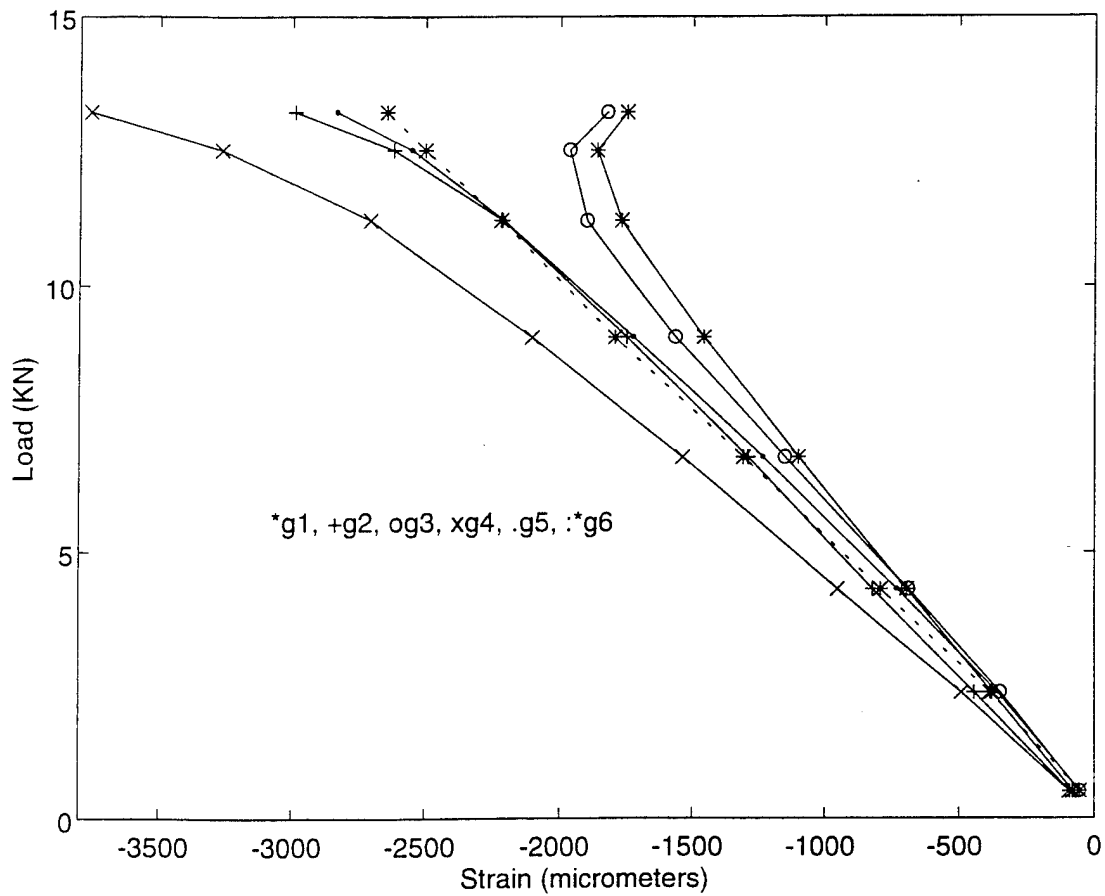


Figure 41. Compression Test, 12.70 mm Thick, 6.35 mm Dia. Hole at Each Quarter Point.

| load (KN) | gauge1 | gauge2 | gauge3 | gauge4 |
|-----------|--------|--------|--------|--------|
| 0.49 | -57 | -86 | -64 | -136 |
| 1.11 | -132 | -208 | -164 | -307 |
| 2.30 | -299 | -434 | -373 | -600 |
| 3.71 | -492 | -697 | -624 | -925 |
| 0.52 | -707 | -988 | -916 | -1263 |
| 7.35 | -997 | -1393 | -1323 | -1729 |
| 9.03 | -1222 | -1726 | -1645 | -2109 |
| 10.76 | -1420 | -2104 | -1979 | -2510 |
| 11.84 | -1505 | -2390 | -2195 | -2774 |
| 12.81 | -1474 | -2744 | -2368 | -3040 |
| 13.41 | -1183 | -3183 | -2358 | -3450 |

Table 27. Compression Test, 12.70 mm Thick, One 6.35 mm Dia. Hole at One Quarter Point.

| load (KN) | gauge1 | gauge2 | gauge3 | gauge4 | gauge5 | gauge6 |
|-----------|--------|--------|--------|--------|--------|--------|
| 0.50 | -80 | -92 | -57 | -94 | -53 | -54 |
| 2.36 | -378 | -444 | -347 | -491 | -358 | -386 |
| 4.31 | -693 | -819 | -687 | -951 | -728 | -790 |
| 6.77 | -1100 | -1292 | -1152 | -1536 | -1235 | -1310 |
| 9.04 | -1458 | -1744 | -1563 | -2104 | -1718 | -1789 |
| 11.21 | -1765 | -2215 | -1897 | -2705 | -2211 | -2220 |
| 12.51 | -1859 | -2618 | -1964 | -3265 | -2550 | -2500 |
| 13.21 | -1745 | -2990 | -1820 | -3759 | -2833 | -2643 |

Table 28. Compression Test, 12.70 mm Thick, 6.35 mm Dia. Hole at Each Quarter Point.

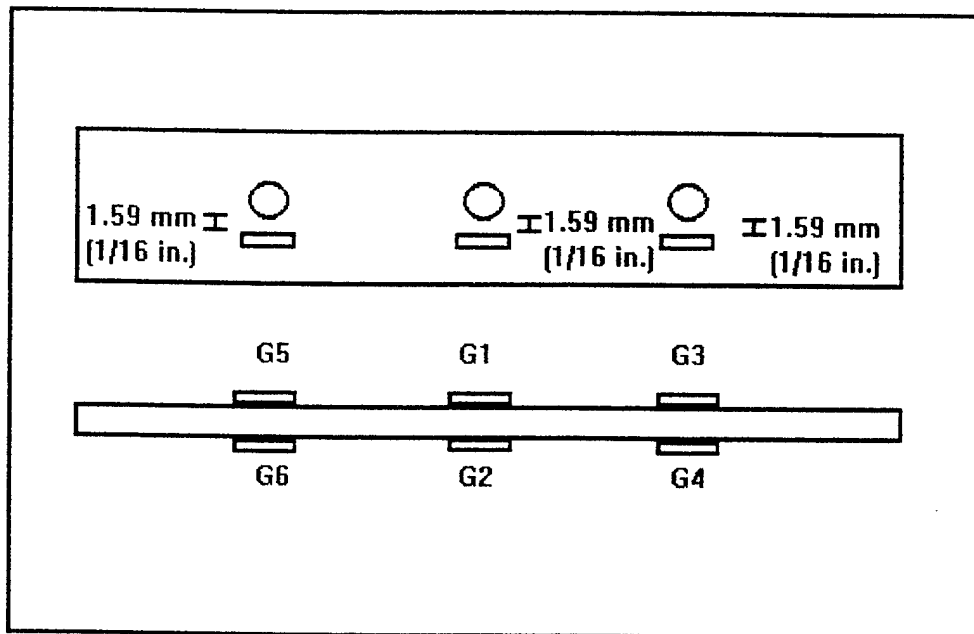


Figure 42. Strain Gauge Locations for Samples with Holes at Each Quarter Point and at the Center.

| load (KN) | gauge1 | gauge2 | gauge3 | gauge4 | gauge5 | gauge6 |
|-----------|--------|--------|--------|--------|--------|--------|
| 0.93 | -274 | -144 | -257 | -165 | -302 | -122 |
| 2.18 | -561 | -388 | -510 | -433 | -629 | -352 |
| 3.81 | -898 | -728 | -809 | -811 | -1014 | -666 |
| 6.43 | -1453 | -1273 | -1324 | -1402 | -1617 | -1195 |
| 9.15 | -2056 | -1811 | -1872 | -2015 | -2263 | -1722 |
| 11.55 | -2624 | -2285 | -2342 | -2600 | -2896 | -2183 |
| 12.42 | -2821 | -2465 | -2469 | -2854 | -3081 | -2366 |
| 13.15 | -2939 | -2689 | -2510 | -3165 | -3256 | -2521 |

Table 29. Compression Test, 12.70 mm Thick, 6.35 mm Dia. Hole at Each Quarter Point and at the Center.

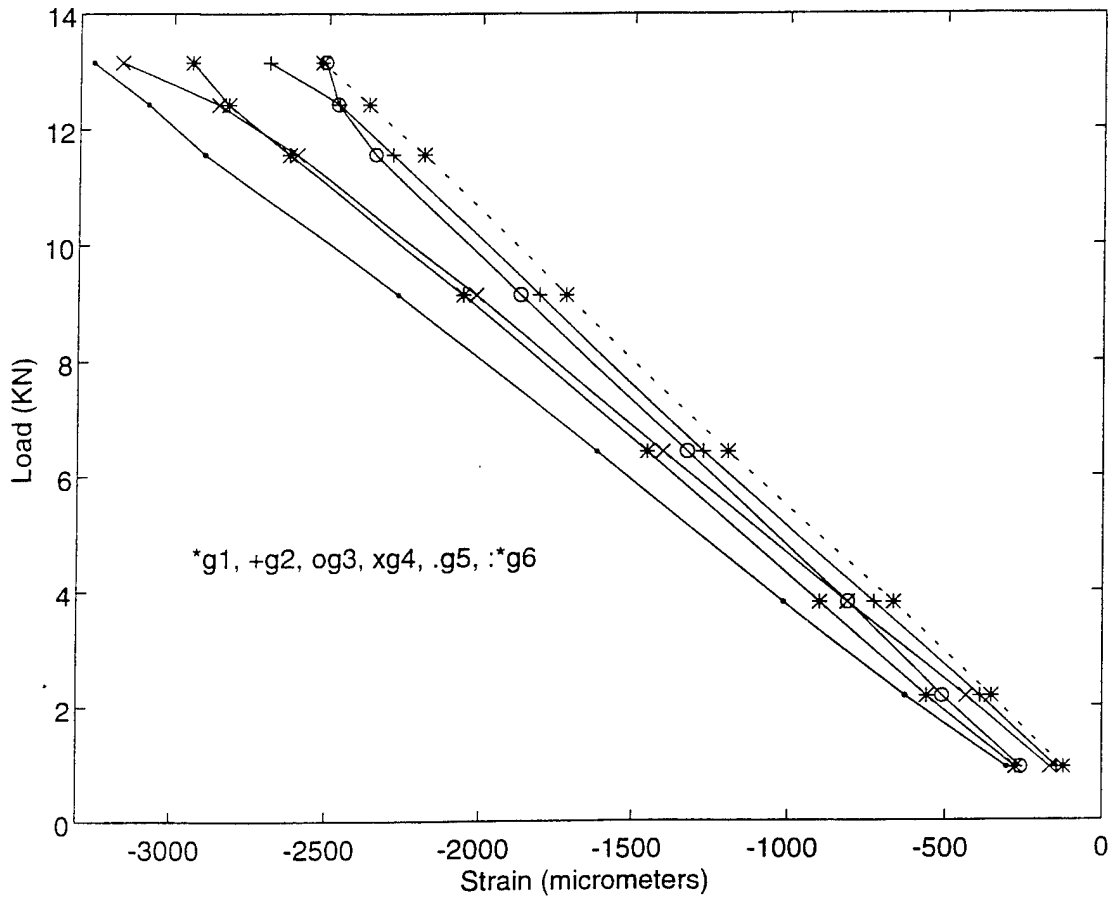


Figure 43. Compression Test, 12.70 mm Thick, 6.35 mm Dia. Hole at Each Quarter Point and at the Center.

| | 3 mm thick foam | 6.35 mm thick foam | 12.70 mm thick foam |
|--|------------------------|---------------------------|----------------------------|
| no hole | 3.47 KN at qtr pt | 6.32 KN at qtr pt | 14.23 KN at qtr pt |
| 6.35mm hole at the center | | | 12.31 KN at qtr pt |
| 12.70 mm hole at the center | 2.74 KN at qtr pt | 6.13 KN at qtr pt | 10.23 KN at center |
| 19.05 mm hole at the center | | 5.24 KN at center | 10.28 KN at center |
| 22.23 mm hole at the center | 2.61 KN at center | | |
| 6.35 mm hole at one quarter point | | | 13.41 KN at qtr pt |
| 6.35 mm hole at both quarter points | | | 13.21 KN at qtr pt |
| 6.35 mm hole at both quarter points and at center | | | 13.15 KN at qtr pt |
| 12.70 mm hole at both quarter points | | 5.74 KN at qtr pt | |
| 19.05 mm hole at both quarter points | | 4.99 KN at qtr pt | |

Table 30. Compression Test Summary Chart.

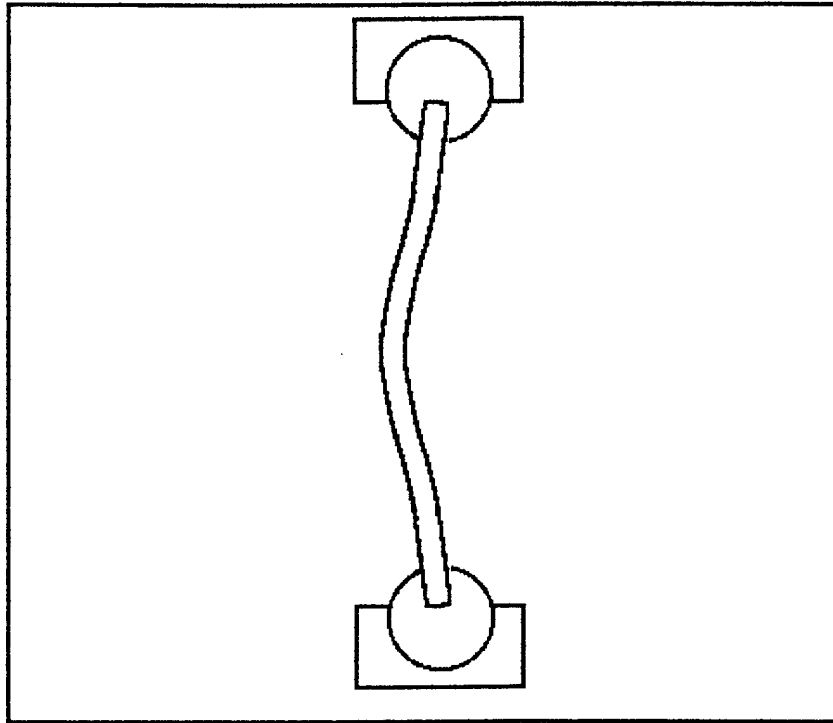


Figure 44. Sample Shape Characteristic of Compression Test.

VI. COMPRESSION WITH DELAMINATION

Four compression tests on delaminated samples were conducted. All four samples had a foam thickness of 6.35 mm (0.25 in.) with a 12.70 mm (0.5 in.) hole at one quarter point. Strain gauges were mounted as shown in Figure 39. Results were compared to the no-hole results obtained from Clawson's study [Ref. 2]. In all cases the delamination was on one side only and was centered length-wise.

The first sample had 101.6 mm (4.0 in.) of delamination and failed by further delamination at a load of 1.1 KN. The failure load was approximately the same as the no-hole failure load obtained by Clawson [Ref. 2]. The results are shown in Figure 45 and Table 31.

The second sample had 50.8 mm (2.0 in.) of delamination and failed by further delamination at a load of 2.7 KN. The failure load was the same as the failure load for the no-hole case. See Figure 46 and Table 32.

The third sample had 25.4 mm (1.0 in.) of delamination and failed at the hole at a load of 4.96 KN. The failure load was 83 percent of the no-hole failure load. Figure 47 and Table 33 shows the results.

The fourth sample had 12.70 mm (0.5 in.) of delamination and failed at the hole at a load of 7.12 KN. The failure load was 93 percent of the no-hole failure load. See Figure 48 and Table 34.

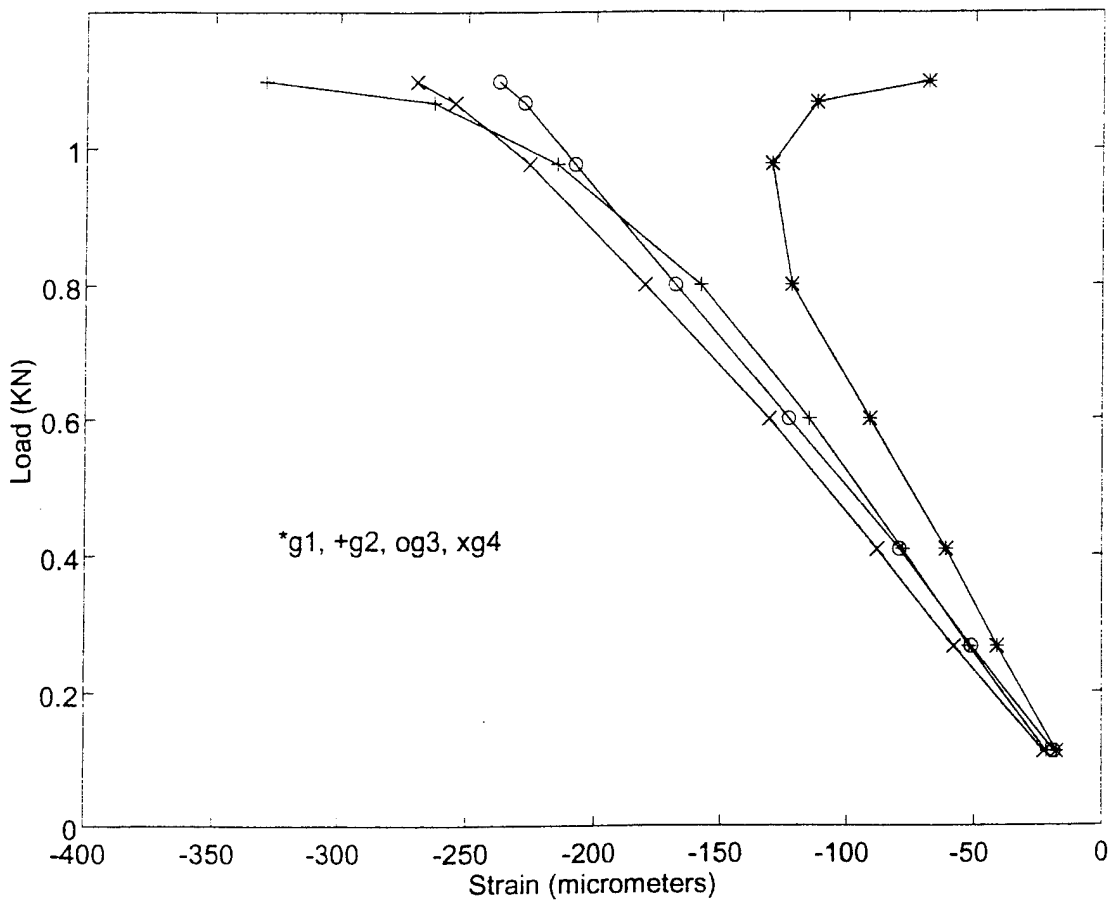


Figure 45. 6.35 mm Thick, 101.6 mm Delamination, 12.70 mm Hole at Quarter Point.

| load (KN) | gauge1 | gauge2 | gauge3 | gauge4 |
|-----------|--------|--------|--------|--------|
| 0.11 | -18 | -22 | -19 | -23 |
| 0.27 | -41 | -52 | -51 | -58 |
| 0.41 | -61 | -78 | -79 | -88 |
| 0.60 | -91 | -115 | -123 | -131 |
| 0.80 | -122 | -158 | -168 | -180 |
| 0.98 | -130 | -215 | -208 | -226 |
| 1.07 | -112 | -263 | -228 | -255 |
| 1.10 | -68 | -330 | -238 | -270 |

Table 31. 6.35 mm Thick, 101.6 mm Delamination, 12.70 mm Hole at Quarter Point.

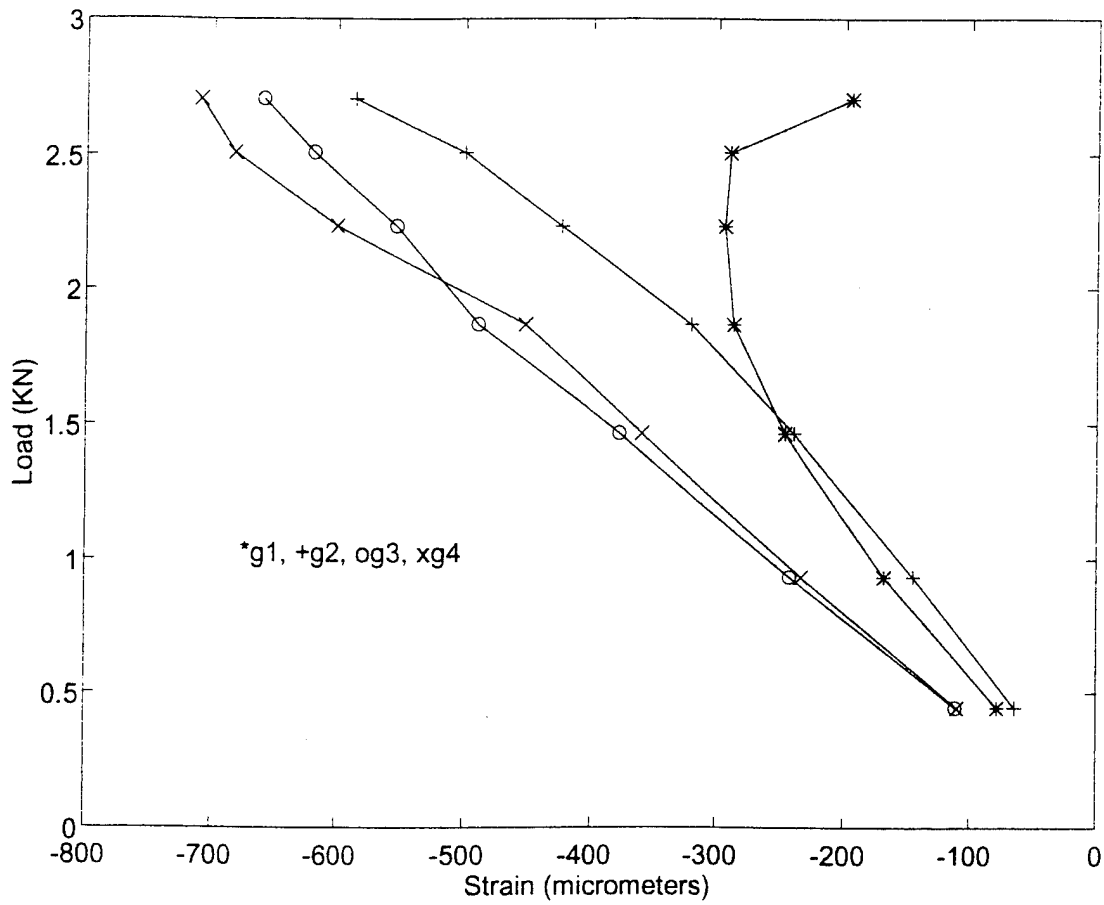


Figure 46. 6.35 mm Thick, 50.8 mm Delamination, 12.70 mm Hole at Quarter Point.

| load (KN) | gauge1 | gauge2 | gauge3 | gauge4 |
|-----------|--------|--------|--------|--------|
| 0.44 | -78 | -64 | -111 | -110 |
| 0.93 | -168 | -145 | -242 | -233 |
| 1.47 | -246 | -239 | -377 | -359 |
| 1.87 | -287 | -320 | -489 | -452 |
| 2.24 | -294 | -424 | -553 | -600 |
| 2.51 | -290 | -500 | -619 | -683 |
| 2.70 | -195 | -586 | -660 | -710 |

Table 32. 6.35 mm Thick, 50.8 mm Delamination, 12.70 mm Hole at Quarter Point.

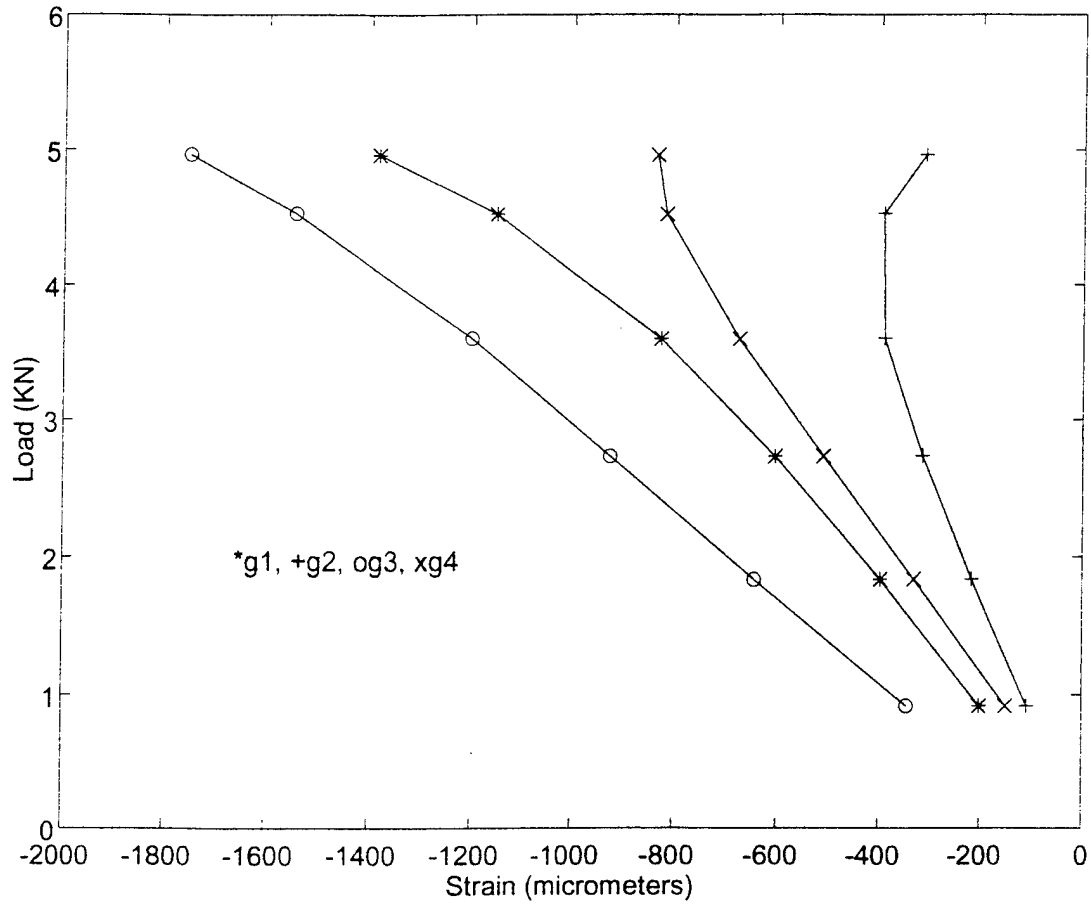


Figure 47. 6.35 mm Thick, 25.4 mm Delamination, 12.70 mm Hole at Quarter Point.

| load (KN) | gauge1 | gauge2 | gauge3 | gauge4 |
|-----------|--------|--------|--------|--------|
| 0.92 | -201 | -108 | -344 | -150 |
| 1.84 | -397 | -217 | -644 | -331 |
| 2.73 | -604 | -315 | -923 | -510 |
| 3.60 | -826 | -391 | -1200 | -676 |
| 4.53 | -1153 | -393 | -1545 | -817 |
| 4.96 | -1384 | -311 | -1752 | -835 |

Table 33. 6.35 mm Thick, 25.4 mm Delamination, 12.70 mm Hole at Quarter Point.

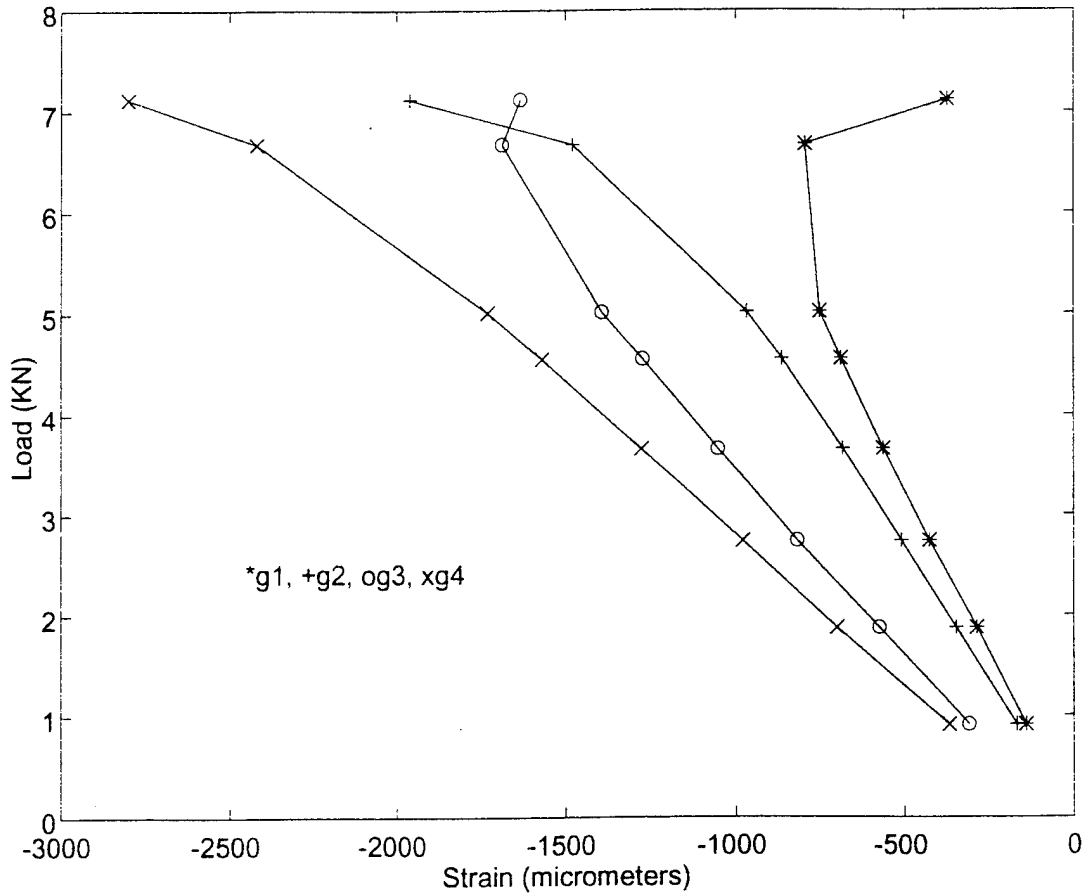


Figure 48. 6.35 mm Thick, 12.70 mm Delamination, 12.70 mm Hole at Quarter Point.

| load (KN) | gauge1 | gauge2 | gauge3 | gauge4 |
|-----------|--------|--------|--------|--------|
| 0.92 | -140 | -167 | -307 | -367 |
| 1.88 | -284 | -347 | -572 | -700 |
| 2.74 | -425 | -507 | -817 | -980 |
| 3.66 | -562 | -682 | -1051 | -1275 |
| 4.56 | -688 | -863 | -1271 | -1568 |
| 5.02 | -751 | -968 | -1390 | -1730 |
| 6.67 | -794 | -1477 | -1686 | -2419 |
| 7.12 | -372 | -1960 | -1632 | -2800 |

Table 34. 6.35 mm Thick, 12.70 mm Delamination, 12.70 mm Hole at Quarter Point.

VII. FINITE ELEMENT ANALYSIS

A finite element analysis was conducted to better understand the experimental results. The ANSYS Engineering Analysis System revision 4.4 was used to conduct the numerical modeling. Five models were constructed representing the no-hole case, the center hole case, the quarter point hole case, the case with a hole at each quarter point, and the case with a hole at each quarter point and at the center. Figures 49, 50, 51, 52, and 53 show the element mesh.

A linear buckling analysis was conducted first. Smeared mechanical properties for the composite were calculated and a four-node shell element was used to model the specimens. The numerical results are shown in Table 35 in conjunction with the experimental results. For the samples with core thicknesses of 3 mm (0.12 in.) and 6.35 mm (0.25 in.), the experimental failure loads are consistently higher than the numerical linear buckling loads. This can be explained by friction in the experimental boundary fixtures. The friction caused the actual experimental boundary conditions to be less than ideal freely rotating end conditions, which caused the experimental results to be higher than the numerical results. The numerical results were therefore normalized to the no-hole case and are summarized in Table 36 in conjunction with the normalized experimental results. For the samples with a core thickness of 12.70 mm (0.5 in.), the higher loads overcame the friction and the end fixtures behaved more like simply supported boundary conditions. Therefore the numerical results agreed with the experimental results.

The friction at the end support can be modeled by moments applied at the boundaries as shown in Figure 54. The Governing Equation is

$$\frac{d^2v}{dx^2} + k^2v = -\frac{1}{EI} M_o \quad \dots\dots\dots (1)$$

where

$$k = \sqrt{\frac{P}{EI}} \quad \dots\dots\dots (2)$$

The deflection equation is

$$v = -\frac{M_o}{P} \frac{(\cos \sqrt{\frac{P}{EI}} L - 1)}{\sin \sqrt{\frac{P}{EI}} L} \sin \sqrt{\frac{P}{EI}} x + \frac{M_o}{P} (\cos \sqrt{\frac{P}{EI}} x - 1) \quad \dots\dots\dots (3)$$

The Solution for the shear force V is

$$V = \frac{M_o k}{\cos \frac{kL}{2}} \sin \left(\frac{kL}{2} - kx \right) \quad \dots\dots\dots (4)$$

Algebraic rearrangement of this expression results in the equation given in reference 7.

Shear force V is a maximum when

$$\frac{kL}{2} - kx = \frac{\Pi}{2} \quad \dots\dots\dots (5)$$

Solving for x

$$x = \frac{L}{2} \left(1 - \sqrt{\frac{P_{cr}}{P}} \right) \dots\dots\dots (6)$$

Where P is the failure load and P_{cr} is the Euler buckling load.

$$P_{cr} = \frac{\Pi^2 EI}{L^2} \dots\dots\dots (7)$$

The sample with a core thickness of 3 mm (0.12 in.) has a failure load of 3.47 KN and a Euler buckling load of 1.29 KN. For this sample the analysis yields x = 0.2L. The sample with a core thickness of 6.35 mm (0.25 in.) has a failure load of 6.32 KN and a Euler buckling load of 4.18 KN. For this sample the analysis yields x = 0.1L. The experimental results for these two samples show good correlation to the predicted failure locations.

Continuing to solve for the bending moment M

$$M = \frac{M_o}{\cos \frac{kL}{2}} \cos \left(kx - \frac{kL}{2} \right) \dots\dots\dots (8)$$

The maximum bending moment occurs at

$$x = \frac{L}{2} \dots\dots\dots (9)$$

Examination of the experimental samples shows that failure at the hole is due to bending while failure at the quarter point away from the hole is due to foam core shear failure. A geometric non-linear stress analysis using an eight-node layered shell element was conducted to take a closer look at the stresses within the different composite layers to support the experimental findings. The analysis shows that a sample without a hole subjected to a given compressive load has a greater foam core shear stress at the quarter point than at the center. With a hole at the center, the foam core shear stress at the quarter point increases as the hole diameter increases. However, the carbon skin bending stress at the hole increases at a faster rate than the foam core shear stress. When the hole diameter reaches a critical diameter, the bending stress at the hole becomes more critical in failure than the core shear stress at the quarter point. As a result, the failure mode changes from foam core shear failure at the quarter point to skin bending failure at the hole. Figure 55 shows the rate of increase for both stresses with increasing hole diameter. With a 6.35 mm (0.25 in.) hole, the sample fails at the quarter point due to foam core shear failure. With a 12.70 mm (0.5 in.) hole, the bending stress at the hole is more critical than the core shear stress at the quarter point and the sample fails at the hole. Between these two points, the failure mode makes a transition from foam core shear failure at the quarter point to bending failure at the hole. Further refinement of the transition location is a recommended area for further study.

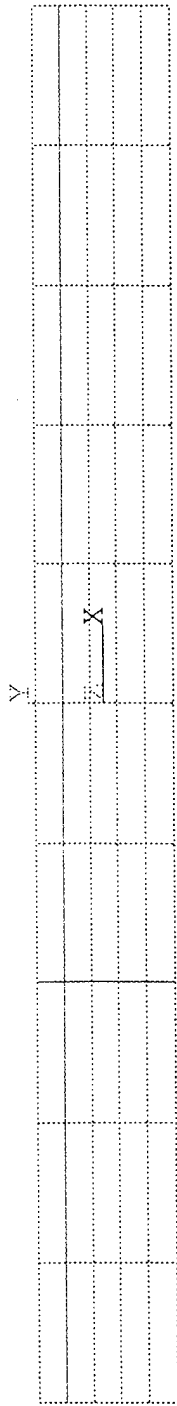


Figure 49. Finite Element Mesh for the No-Hole Case.

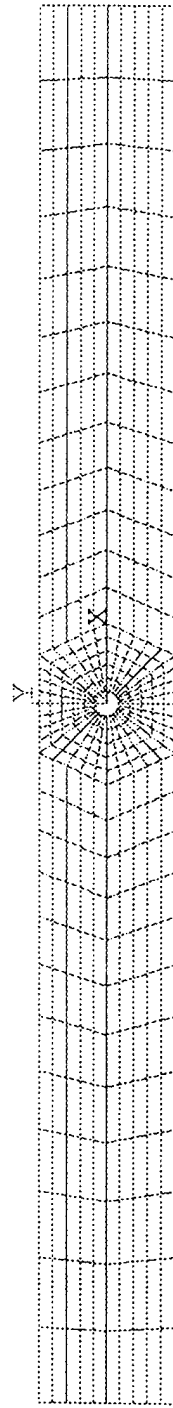


Figure 50. Finite Element Mesh for the Center Hole Case.

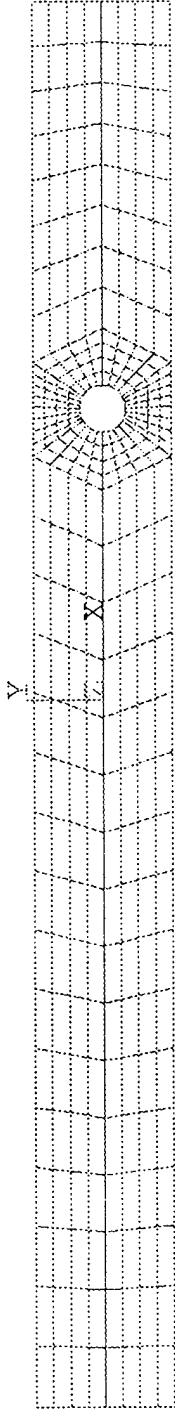


Figure 51. Finite Element Mesh for the Quarter Point Hole Case.

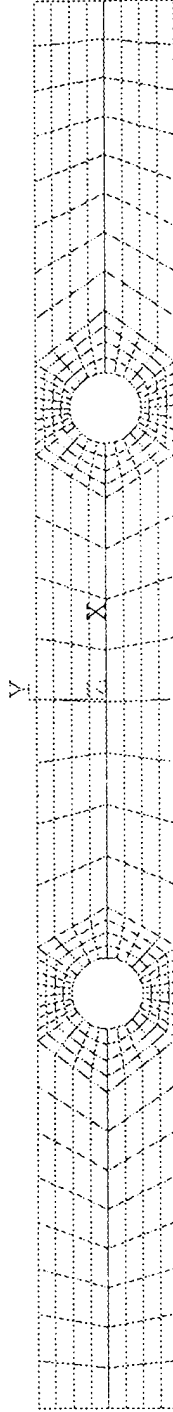


Figure 52. Finite Element Mesh for the Case with a Hole at Each Quarter Point.

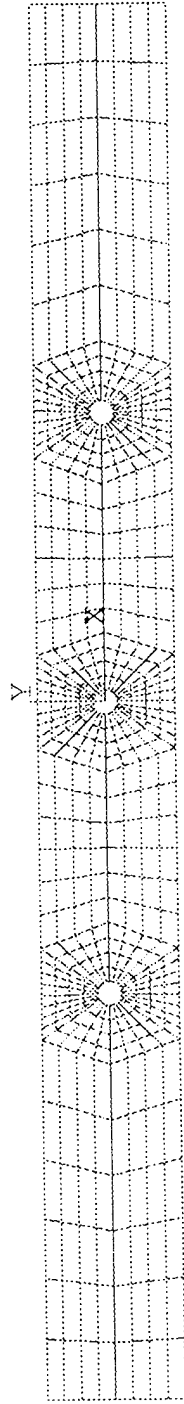


Figure 53. Finite Element Mesh for the Case with Holes at Both Quarter Points and at the Center.

| Key: Experimental Numerical | 3 mm thick foam | 6.35 mm thick foam | 12.70 mm thick foam |
|--|------------------------------|------------------------------|--------------------------------|
| no hole | 3.47 KN at qtr pt 1.29 KN | 6.32 KN at qtr pt 4.18 KN | 14.23 KN at qtr pt 14.31 KN |
| 6.35mm hole at the center | 1.27 KN | 4.14 KN | 12.31 KN at qtr pt 14.18 KN |
| 12.70 mm hole at the center | 2.74 KN at qtr pt 1.24 KN | 6.13 KN at qtr pt 4.04 KN | 10.23 KN at center 13.86 KN |
| 19.05 mm hole at the center | 1.16 KN | 5.24 KN at center 3.82 KN | 10.28 KN at center 13.17 KN |
| 22.23 mm hole at the center | 2.61 KN at center 1.12 KN | 3.67 KN | 12.65 KN |
| 6.35 mm hole at one quarter point | 1.27 KN | 4.15 KN | 13.41 KN at qtr pt 14.24 KN |
| 6.35 mm hole at both quarter points | 1.27 KN | 4.13 KN | 13.21 KN at qtr pt 14.17 KN |
| 6.35 mm hole at both quarter points and at center | 1.26 KN | 4.10 KN | 13.15 KN at qtr pt 14.16 KN |
| 12.70 mm hole at both quarter points | 1.20 KN | 5.74 KN at qtr pt 4.01 KN | 13.78 KN |
| 19.05 mm hole at both quarter points | 1.13 KN | 4.99 KN at qtr pt 3.74 KN | 12.91 KN |

Table 35. Numerical and Experimental Result.

| Key: Experimental Numerical | 3 mm thick foam | 6.35 mm thick foam | 12.70 mm thick foam |
|--|----------------------------|-------------------------------|--------------------------------|
| no hole | 1.0 at qtr pt 1.0 | 1.0 at qtr pt 1.0 | 1.0 at qtr pt 1.0 |
| 6.35mm hole at the center | 0.99 | 0.99 | 0.87 at qtr pt 0.99 |
| 12.70 mm hole at the center | 0.79 at qtr pt 0.97 | 0.97 at qtr pt 0.97 | 0.72 at center 0.97 |
| 19.05 mm hole at the center | 0.91 | 0.83 at center 0.91 | 0.72 at center 0.92 |
| 22.23 mm hole at the center | 0.75 at center 0.87 | 0.88 | 0.88 |
| 6.35 mm hole at one quarter point | 0.99 | 0.99 | 0.94 at qtr pt 0.99 |
| 6.35 mm hole at both quarter points | 0.99 | 0.99 | 0.93 at qtr pt 0.99 |
| 6.35 mm hole at both quarter points and at center | 0.98 | 0.98 | 0.92 at qtr pt 0.99 |
| 12.70 mm hole at both quarter points | 0.96 | 0.91 at qtr pt 0.96 | 0.96 |
| 19.05 mm hole at both quarter points | 0.89 | 0.79 at qtr pt 0.90 | 0.90 |

Table 36. Normalized Experimental and Numerical Results.

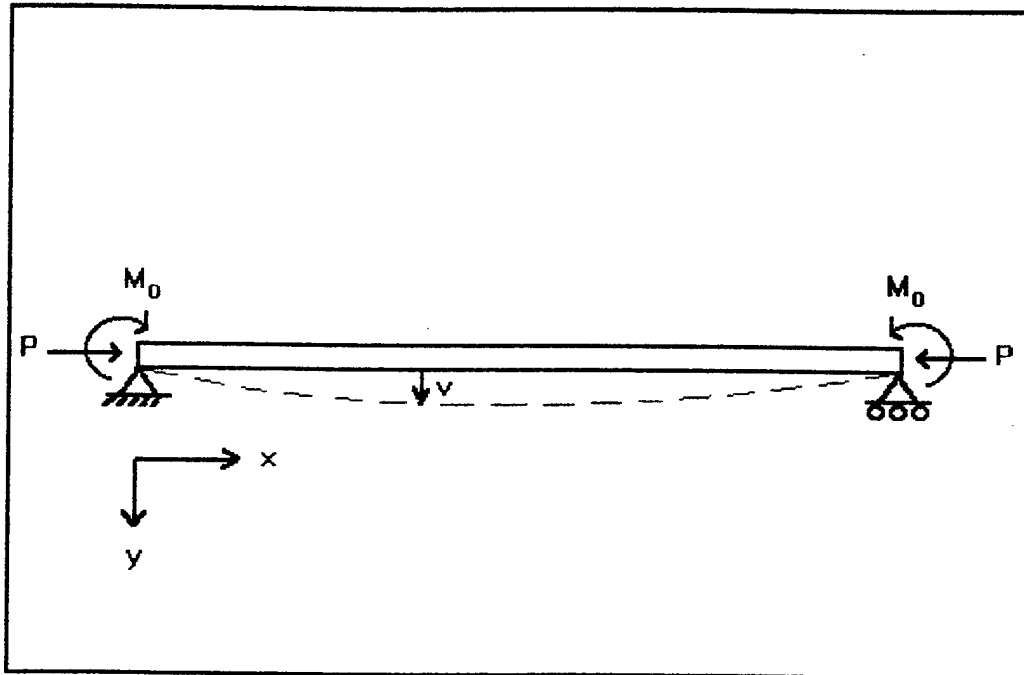


Figure 54. End fixture friction model.

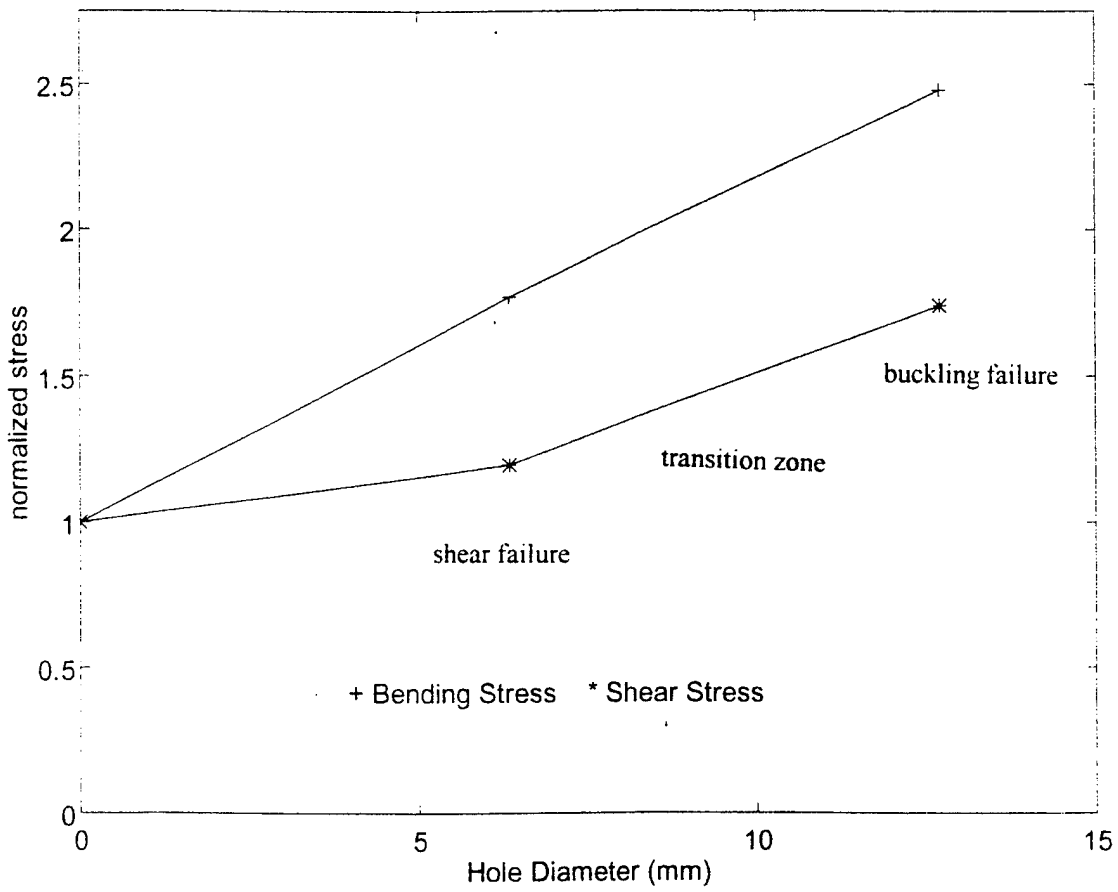


Figure 55. Normalized stresses vs. Hole Diameter.

VIII. CONCLUSIONS AND RECOMMENDATIONS

Four-point bending is not an effective test method to evaluate the effects of stress concentration at a hole in a foam cored sandwich composite beam as foam core shear failure occurs at a support point prior to bending failure at the hole.

A foam cored sandwich composite beam loaded in compression has a greater foam core shear stress between the end support and the quarter point than at the center and fails between the end support and the quarter point by foam core shear failure. With a hole at the center, the foam core shear stress at the quarter point increases as the hole diameter increases. However, the carbon skin bending stress at the hole increases at a faster rate than the foam core shear stress. When the hole diameter reaches a critical diameter, the bending stress at the hole becomes more critical in failure than the core shear stress at the quarter point. As a result, the failure mode changes from foam core shear failure at the quarter point to skin bending failure at the hole. The critical diameter decreases as the foam core thickness increases. Sandwich composites with a thick core are most commonly used in practical applications. When thick core composites are used, the critical diameter is small and failure is by bending.

The experimental compression failure loads are higher than the numerical linear buckling loads for samples with core thicknesses of 3 mm (0.12 in.) and 6.35 mm (0.25 in.). This can be explained by friction in the experimental boundary fixtures. The friction at the end supports can be modeled by moments applied at the boundaries. As derived in Chapter

VII, the failure location can be calculated.

An area of future study is the interaction of holes and delamination during the failure process. It is recommended that simply-supported experimental boundary fixtures with low friction be developed.

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