

# AIR CORPS INFORMATION CIRCULAR

(AVIATION)

PUBLISHED BY THE CHIEF OF AIR CORPS, WASHINGTON, D. C.

Vol. VI

May 21, 1927

No. 586

## STUDY OF BALANCED RUDDERS



Prepared by J. A. Roche  
Matériel Division, Air Corps  
McCook Field, Dayton, Ohio  
February 16, 1927



**EXHIBITION STATEMENT A**  
Approved for public release;  
Distribution Unlimited

19980820 006

UNITED STATES  
GOVERNMENT PRINTING OFFICE  
WASHINGTON  
1927

EQUIPMENT DIVISION

CERTIFICATE: By direction of the Secretary of War the matter contained herein is published as administrative information and is required for the proper transaction of the public business.

(11)

# STUDY OF BALANCED RUDDERS

## OBJECT

The object of this study was to discover to what extent the rudder can be balanced without causing pilots to complain of an overbalanced condition.

## DISCUSSION

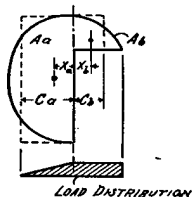
The dimensions of the various rudders in use on military airplanes were collected and an expression was developed to show the relative extent to which these rudders are balanced. The development of this expression assumed a uniform load distribution on the balanced portion and a gradually decreasing load from the hinge to the trailing edge. The coefficient  $K_b$  is, therefore, a ratio of the balanced area to the area aft of the hinge, due weight being given to the effectiveness of both areas. Pilots observations on the action of these rudders were also obtained and are given under the heading of "Remarks" in the table below.

## CONCLUSIONS

This study shows no definite relation between the balanced coefficients and the pilots remarks. It may,

therefore, be that rudders which are reported as overbalanced are really not overbalanced, and that other yaw characteristics of the airplane have resulted in producing that impression on the pilot. Again it may be that the shape of the balanced area influences its effectiveness in a way which we are unable to understand, still another possibility is that the area aft of the hinge may be shielded to various extents in the different airplanes by the fuselage, stabilizer, and elevators, also to an extent which we can not quantitatively determine. It is noteworthy that the XO-6 airplane has the smallest balance coefficient and yet is criticized by the pilots for being overbalanced. In view of our further study of the yaw characteristics of this airplane it is fairly safe to conclude that the overbalanced feeling is due to certain irregularities in the yaw moment curves for which either the nose of the airplane or the nose of the fin may be responsible. It is also worthy of note that the DH-4, the rudder characteristics of which are satisfactory, shows the highest value of balance coefficient and that this coefficient is more than three times as great as that of the alleged overbalanced XO-6.

## DATA ON RUDDER BALANCES OF VARIOUS AIRPLANES



- $A_t$  = Total Area of Rudder
- $A_a$  = Area aft of hinge center line
- $A_b$  = Area of balance fore of hinge center line
- $C_a$  = Mean Chord of  $A_a$
- $C_b$  = Mean Chord of  $A_b$
- $X_a = \frac{C_a}{3}$  = Dist. to C. P. of  $A_a$  (approximate)
- $X_b$  = Distance to center of area  $A_b$
- $K_b = \frac{\text{Moment of Load on } A_b}{\text{Moment of Load on } A_a} = \frac{2 A_b X_b}{A_a X_a}$

Airplane	Rudder	Remarks	Reference	$A_t$	$A_a$	$A_b$	$C_a$	$C_b$	$X_a$	$X_b$	$K_b$
CO-6	Original	Overbalanced	Captain Brower	Sq. ft. 14.82	Sq. ft. 12.73	Sq. ft. 2.09	In. 25.50	In. 12.10	In. 8.50	In. 7.02	0.260
CO-6	Reduced	O. K.	do.	14.27	12.73	1.54	25.50	8.88	8.50	4.94	.141
CO-5	Present	O. K. (small $A_b$ ) <sup>2</sup>	Lieutenant Johnson	18.41	16.50	1.91	28.60	15.30	9.53	8.60	.209
CO-4	do.	O. K.	do.	13.70	12.32	1.38	24.95	8.25	8.32	3.93	.106
CO-4	do.	First article	do.	9.35	7.99	1.66	22.15	10.12	7.38	5.24	.306
DH-4	Present	O. K.	do.	12.20	10.40	1.80	23.60	12.50	7.87	7.87	.345
O-1	do.	O. K.	do.	11.80	10.60	1.20	22.00	10.13	7.33	5.94	.184
P-2	do.	O. K.	do.	10.80	9.32	1.48	22.18	16.40	7.39	6.65	.286
A-1 <sup>3</sup>	do.	Overbalanced	Lieutenant Diehman	13.75	11.90	1.85	29.60	7.82	9.87	3.97	.125
	do.	O. K.	Lieutenant Johnson	36.40	30.54	5.86	46.90	20.85	15.63	12.51	.307
GAX-1	Original	Overbalanced	Mr. Laddon	34.00	30.54	3.46	46.90	23.70	15.63	14.20	.206
GAX-1	Reduced	O. K.	do.	13.32	11.98	1.34	28.80	16.05	9.60	8.24	.192
PT-1	Present	Very good	Lieutenant Johnson	17.27	15.80	1.47	26.90	8.39	8.97	4.32	.089
XO-6	do.	Overbalanced	Lieutenant Hutchinson	14.92	13.65	1.27	25.00	9.50	8.33	5.09	.114
XO-6	do.	do.	do.	14.49	12.65	1.84	26.50	14.15	8.83	7.33	.241
XO-6	do.	Suggested change No. 1 by T. M. Co									
XO-6	do.	Change suggested by McCook Field									

<sup>1</sup> Triangular distribution of load is assumed for  $A_a$ . Uniform distribution is assumed for  $A_b$ .

<sup>2</sup> Insufficient total area for control at low speeds.

<sup>3</sup> Ambulance.

