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BAA98-02: AERO-STRUCTURAL INTERACTION AND CONTROL

PHASE I FINAL REPORT

On

**ACTIVE FLOW CONTROL ON HURLEY'S FREE-STREAMLINE
AIRFOIL AND DELTA WING**

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ABSTRACT

This was a joint effort by The University of Arizona (UA) and Arizona State University (ASU). The work was directed toward the application of active flow control to a free-streamline airfoil whose structural vibrations were integrated into the control strategy. The Phase I accomplishments reported here demonstrate that efficient free-shear-layer control is possible and the low drag of the free-streamline airfoil is a reality.

1 RESEARCH GOALS

1.1 Introduction

Slender delta wings rely on reattached vortices to provide the necessary lift at low speeds and high angles of attack. Without some form of control, when the angle of attack exceeds a certain threshold value on an ordinary delta wing, the vortex breaks down and lifts off the surface (Fig. 1a). This process results in a loss of lift and an abrupt increase in drag. Many attempts, therefore, have been made (e.g., vortex flaps) to eliminate vortex breakdown or at least to delay it. When periodic excitation was provided at the leading edge of this delta planform, the vortex did not separate from the surface; the location of the reattachment line was detected to remain parallel to the leading edge (Fig. 1b). One may extrapolate the results obtained for the "generic-flap" (Nishri et al. 1998) and the swept-back wing (Naveh et al. 1998) to a finite-wing planform. They suggested that periodic forcing applied to a delta wing over which the flow was separated should cause reattachment scaling with the dimensionless frequency $F_{3D}^+ = f^*X_d/U_N$, where X_d is the distance between the leading-edge and the reattachment location *normal* to the leading edge, f is the forcing frequency, and $U_N = U_\infty \cos \Lambda$ is the *normal* component of velocity. Although this assumption appeared to be tenuous (it was based on a transformation relying on an infinitely yawed wing section), it agreed with experiment in the vicinity of the apex or the wing tip. Thus, the reattachment line on the wing surface was parallel to the leading edge except near the apex (Fig. 1b). The result was similar to the flow visualization carried out on the yawed wing.

The free-streamline airfoil proposed by Hurley (1961) is particularly attractive in its possible application to slender delta wings in that it may capture the vortex between the main element and the upstream facing flap (Fig. 2) providing an even larger lift than was hitherto possible with just periodic blowing. Other direct applications to uncrewed aerial vehicles (UAVs), low-speed flight of transonic and supersonic aircraft, and control of complicated inlet flows are also implied and straightforward.

1.2 Accomplishments during Phase I

During Phase I of this proposal, we verified the feasibility of using the free-streamline airfoil as a high-lift device and demonstrated the ability to calculate the flowfield and the structural vibrations. The proposed control strategies worked and the agreement between measurements and computations was very good.

Control of the flow over Hurley's free streamline airfoil

We first repeated the experiment of Hurley on a similar configuration (Fig. 3). The geometrical parameters include the chord c , the flap angle τ , the flap length kc , the angle of incidence of the main element α , and the slot location. The flow parameters include the chord Reynolds number Re , those associated with the control implemented on the flap (steady blowing amplitude $C_{\mu 2}$, periodic blowing rms amplitude $\langle c_{\mu 2} \rangle$, and dimensionless periodic blowing frequency F_2^+), and those associated with the flaperon implemented on the leading edge of the main element (periodic frequency F_1^+ and periodic rms amplitude $\langle c_{\mu 1} \rangle$).

As mentioned in the Introduction above, previous research has shown that the *normal-to-the-leading-edge* component contains the relevant physics. The slot in the present experiment was moved to enable the oscillations and the steady blowing to emanate from the gap between the circular cylinder and the upper surface (Fig. 3). The basic airfoil provided a maximum lift of 1.4 at an incidence of 20° (Fig. 4). Application of steady blowing increased the maximum lift particularly when the applied momentum coefficients $C_{\mu 2}$ was greater than 5%. At this $C_{\mu 2}$ a C_{Lmax} of 2.7 was achieved (representing double the base value) while the drag at $C_L = 1$ was halved (Fig. 4). The application of steady blowing to reattach the flow to the solid surface is very inefficient because it relies on the natural entrainment of external fluid by a high-speed wall jet. It thus requires a relatively large $C_{\mu 2}$.

Since it is well known that an oscillatory motion can keep the flow attached or can force the reattachment of the flow much more efficiently than steady blowing (Seifert et al. 1993, 1996), a time-periodic blowing was used in this case. The data presented in Fig. 5 indicate that periodically exciting the jet at $\langle c_{\mu 2} \rangle = 0.5\%$ achieves approximately the same increase in the lift as that attained earlier with steady blowing of $C_{\mu 2} = 5\%$. Thus the maximum lift coefficient can be doubled by periodic excitation that uses only 10% of the momentum needed to achieve the same result via steady blowing. This improvement in C_{Lmax} required an optimization of the reduced frequency, F_2^+ , which occurred when $F_2^+ = 3$. This value of F_2^+ is consistent with previous experiments aimed at delaying separation (Nishri et al. 1998). Adding the same value of $\langle c_{\mu 2} \rangle = 0.5\%$ at $F_2^+ = 1.1$ generated a small increase in C_{Lmax} less than 20% (Fig. 6) and did not decrease the drag. Combining steady blowing with periodic excitation increased the maximum lift at proportionally smaller increments. For example, the addition of $C_{\mu 2} = 2\%$ to the existing excitation of $\langle c_{\mu 2} \rangle = 0.5\%$ increased C_{Lmax} from 2.7 to 3.05, while the addition of $C_{\mu 2} = 5\%$ raised C_{Lmax} to 3.25 (Fig. 7). At small angles of incidence, the addition of steady blowing sufficed to control separation making periodic excitation superfluous. However at α greater than 15° any addition of oscillations was beneficial provided $C_{\mu 2}$ was less than 10%. Oscillations emanating from the slot reduced the pressure upstream of the cylinder (on the upper surface of the main element; see Fig. 3), changing the average C_p there from -1.7 to -4 . It required ten times larger $C_{\mu 2}$ to achieve a similar reduction in C_p (Fig. 8). One observes from the pressure distribution that purely periodic oscillations are able to attach the flow to the upper surface of the flap.

Oil flow over a thin surface that was inserted perpendicular to the flap enabled flow visualization on the free-streamline airfoil. The patterns observed in Fig. 9 were taken at $\alpha = 15^\circ$ in the presence of periodic oscillations at $F_2^+ = 3$ and $\langle c_{\mu 2} \rangle = 0.5\%$ or steady blowing at $C_{\mu 2} = 7\%$. The visualization was done at the same Re as the quantitative measurements. The velocity in the region bound by the free streamline and the airfoil is negligible with the exception of the vortex that is situated near the leading edge of the flap. The size of this vortex is relatively small (about 25% of the distance between the leading edge of the main element and the leading edge of the flap) yet it continuously provides fluid that is later entrained by the shear layer. Thus the stream in the narrow gap between the flap leading edge and the vortex represents the total fluid entrained from this confined region. It is quite possible that the curvature stabilizes the mixing layer preventing the generation of streamwise vortices. A detailed stability analysis would help to understand and control this flow.

The velocity outside the dividing streamline is fast, as indicated by the streamers of oil visible in the figure. A comparison between steady blowing and periodic excitation indicates that the dividing streamline is located higher when the flow is periodically excited than when the steady blowing fourteen times stronger is employed. This proves the efficacy of periodic excitation in trapping the vortex upstream. The flow pattern also suggests that the leading edge of the flap should perhaps be concave to accommodate the trapped vortex in a more effective manner.

The flow pattern is very different at $\alpha = 0^\circ$, even when the periodic excitation is applied (Fig. 10a). The streamlines are parallel to the lower surface near its leading edge. The flow separates slightly ahead of the flap leading edge with the separation streamline normal to the lower surface. No trapped vortex is visible in the gap between the flap and the lower surface. However, streamlines located above the flap are highly curved near the slot. This implies that a very low pressure should exist at the leading edge of the flap while C_p should be zero near the leading edge of the main element. The corresponding pressure distributions with and without periodic excitation of the flow are shown in Fig. 10b. In the absence of any boundary-layer control, the flow is not only separated from the upper surface of the flap but also stagnates in the gap between the flap and the main element. It may be observed (Fig. 10b) that $C_p \cong 1$ on the upper surface of the main element near its leading edge.

The main reason for the efficacy of the pulsating blowing/suction in controlling separation or enhancing mixing is the use of the natural instabilities existing in the mean flow to further amplify the input perturbations. It was known (and the finite element LES calculations confirmed it; see Fig. 11a) that the separation streamline is a curved mixing layer that is naturally unstable at very small Reynolds numbers. Consequently, a periodic excitation (F_1^+) added near its origin at the leading edge of the main element could be used to enhance the instability and increase the entrainment capacity of the mixing layer. For this reason, a small flaperon was also placed near the leading edge of the main element. The harmonic motion of the flaperon (F_1^+ , $\langle c_{\mu 1} \rangle$) provided the necessary excitation to the mixing layer. Initial results suggest that this excitation

could further increase the lift generated by the free-streamline airfoil because it lowered the pressure within the trapped vortex (Fig. 11b). A proper parametric study of this effect will be conducted during the second phase of this project.

Fluid-Structure Interaction

As part of the first phase of this project, a coupled fluid-structure interaction analysis algorithm was developed based on the Finite Element Method (FEM). As described in Fig. 12a-e, this algorithm solves the equations of aerodynamics and structural dynamics in a sequentially coupled manner. The results from the solution of Navier-Stokes equations provide the fluid forces as input for the subsequent structural dynamic analysis. Because the fluid flow and thus the fluid forces are dependent on the geometry of the structure, the computational mesh is updated for the subsequent aerodynamic analysis based on the deformed configuration.

First, the fluid flow in the domain surrounding the structure is calculated by solving the Navier-Stokes equations. A view of a typical computational domain for the fluid flow analysis is shown in Fig. 12a. The flexible structure, in this case a wing, is discretized using shell elements. A view of a typical computational discretization for a wing is shown in Fig. 12b. The pressure distribution obtained from the fluid flow analysis is transferred to the wing as illustrated in Fig. 12c. These forces will cause the wing to deform as shown in Fig. 12d. Because the wing experiences deformation, its shape changes; thus, the fluid flow also changes and the flowfield analysis must be redone. The deformed structural computational mesh for the wing becomes the new boundary positions of the computational domain for the flowfield analysis shown in Fig. 12e. When assigning new boundary positions for the computational domain for the flowfield, the computational mesh must be updated with a re-zoning algorithm. Since the flowfield is highly unsteady, the flow-induced displacements are analyzed in the time domain. Therefore, the above-described steps, as summarized in Fig. 12f, must be performed at every time step of the calculation.

The structural-analysis part of the algorithm developed by Madenci and Barut (1996) and later applied to moderately thick composite components by Barut et al. (1997) accounts for the coupling between the rigid-body motion and elastic deformations. Unlike others, this algorithm is capable of capturing the motion-induced membrane and stiffness variations of general shell structures because the nonlinear strain measure and rotations are explicitly included in the formulation. The formulation of the mass matrix includes the coupling between rigid-body motion, inertial forces and elastic deformation so that the inertial loads can be determined as part of the solution. Changes in the inertial loads due to rapid maneuver or external aerodynamic and actuator loads are captured in this formulation; they can be significant in the aero-elastic tailoring of the aircraft or aerospace structure. In addition, this algorithm, described in detail by Madenci and Barut (1996), accounts for large elastic displacements and rotations, material anisotropy, transverse shear deformations, motion induced stiffening and softening.

As shown by Madenci and Barut (1996), utilizing concepts from finite-element analysis within the realm of nonlinear theory of elasticity as well as rigid-body dynamics and applying the principle of virtual work in conjunction with co-rotational form of the updated Lagrangian description of motion leads to the coupled and highly nonlinear equations of motion for the finite element. The solution vector provides the position and rigid-body rotation of the body-fixed frame and the incremental total nodal vector that represents the flexible part of the motion. The components of the position and rigid-body rotation vectors are measured with respect to the inertial frame, while the body-fixed frame is chosen to measure the components of the total nodal vector. Therefore, the freedom in the choice of different frames of reference makes this formulation different than other existing formulations in which they use a single frame of reference to measure the overall motion. Relating these vectors is the fully coupled and symmetric mass matrix. In most of the formulations, the terms originating from gyroscopic forces are lumped into the mass matrix in a skew-symmetric form, which results in a non-symmetric mass matrix. However, in this approach, the symmetry of the mass matrix is established by retaining the gyroscopic forces in the right-hand-side vector in addition to the external forces. The vectors arising from the internal forces are also provided to complete the nonlinear forcing terms (all dependent on the solution vector) in the equations of motion. The incremental part of the internal force vector involves the tangential stiffness matrix, which is decomposed as linear and geometric stiffness matrices.

There exists no analysis except for Madenci and Barut (1996) and Barut et al. (1997) regarding the dynamic response of composite shells undergoing large overall motion accompanied by large elastic displacements and rotations. Their analysis is capable of capturing the motion-induced membrane and stiffness variations because the nonlinear strain measure and rotations are explicitly included in the formulation and the inertial forces are determined as part of the solution. In other words, the rigid-body motion is not known a priori. Also, their formulation permits the inclusion of time-dependent external forces such as the aerodynamic loads. It is to be noted that none of the available finite-element analyses, including the commercial programs, have the capability to solve this problem within the realm of multi-body dynamics formalism.

Previous studies relevant to the structural dynamics of a composite component are not capable of including the effect of externally applied dynamic loads such as the aerodynamic forces, time-dependent angular velocity and the influence of the coupling between the rigid-body motion, elastic deformations and the inertial forces. An extensive review of these multi-body dynamics formulations coupled with elastic deformations can be found in a study by Madenci and Barut (1996).

The aerodynamic-analysis part of the algorithm developed by Kjellgren and Hyvarinen (1998) and Kjellgren (1997 and 1998) provides the solution of the unsteady Navier-Stokes equations. The formulation is based on an arbitrary Lagrangian-Eulerian description of motion suitable for moving or deformable bodies. It utilizes a semi-implicit fractional-step finite-element method for general two and three dimensions and a Large Eddy

Simulation (LES) for three-dimensional turbulent flow. The updating of the mesh required between the structural dynamics and aerodynamic analyses is based on a novel re-zoning algorithm.

This FEM-based fluid-structure interaction analysis algorithm applies to both the fluid domain and the structure with complex two- or three-dimensional geometry under general boundary conditions. Also, this algorithm is fully interfaced with the pre-and post processor of ANSYS, a commercially available finite-element program. In order to demonstrate the capability of this algorithm, a number of calculations in both two and three dimensions were performed. The first case is a computational analysis of the structural response due to fluid forces of a three-dimensional Hurley wing with aspect ratio of eight. This calculation was performed at $Re = 1100$. No wing-tip effects were included. The computational domain is shown in Fig. 13a and the computational mesh in Fig. 13b. The pressure caused on the wing by the fluid flow is shown in Figs. 13c and 13d, and the corresponding deformation is shown in Fig. 13e.

The second case is a computational analysis of the flow field around a stationary Hurley wing. This analysis was performed at the actual Reynolds number, $Re = 120,000$, which is high enough for turbulence to play a significant role and thus, a large eddy simulation (LES) turbulence model is included. The analysis is done for a slice of the wing, rendering the calculations three dimensional. A schematic view of the computational domain is shown in Fig. 14a. The mesh contains about 200,000 nodes; a close-up view of a slice of the mesh is shown in Fig. 14b. Velocity and pressure contours are shown in Figs. 14c and 14d. Note that the computational predictions of C_D and C_L are in very close agreement with the experimental measurements and observations.

The third case is the same as the second case except for the presence of the blowing slot for boundary-layer control. Velocity and pressure contours are shown in Figs. 15a and 15b. As observed in these figures, it is clear that the blowing slot promotes reattachment, reduces the wake, and thus improves the performance. From computer animations, it is also clear that the blowing slot keeps the vortex confined between the upper and the lower surfaces. The vortex is stronger and steadier, which also improves the lift characteristics. Overall, the results indicate that the lift and drag coefficients can be improved significantly by boundary-layer control. Again, these computational predictions of C_L compare remarkably well with the experimental measurements and observations.

In the fourth case, in order to demonstrate the capability of the fluid-structure interaction algorithm, the Hurley airfoil with a flexible flap and rather stiff lower surface is calculated at $Re = 2100$. The upper flap is modeled as consisting of two rather flexible quasi-isotropic laminates. A schematic view of the structure also showing the direction of the in-coming flow is shown in Fig. 16a. In order to capture the response over a time interval of the fluid-structure interaction, the time-history of the vertical displacement of the tip of the upper flap is shown in Fig. 16b. Based on this figure, the system seems to be aero-elastically stable. The ve-

locity contours when the upper flap is at its bottom position are shown in Fig. 16c and the velocity contours when it is at the top position are shown in Fig. 16d.

In the fifth case, in addition to elastic deformation, the airfoil is allowed to have rigid-body motion through its attachment to linear and torsional springs at its corner as shown in Fig. 17a. This is a generalization of the "typical section airfoil" (a rigid airfoil attached to springs) often used in aero-elastic analyses and is a very useful model for determining the basic aero-elastic properties of the wing. The time-history of the vertical displacement of the tip of the upper flap is shown in Fig. 17b.

Nonlinear parabolized stability equations

The visualizations shown in the previous sections clearly indicate a natural instability of the shear layer emanating off the main element. The entrainment in the free mixing layer is enhanced by these low-frequency oscillations, resulting in the early roll-up of vortices and more efficient attachment to the flap. To capitalize on this effect, oscillations with frequencies corresponding to those of the disturbances most-amplified naturally are introduced at the leading edge of the lower surface (main element) to further enhance the instabilities. A stability analysis to determine the physics of the free-shear-layer behavior and the disturbances most amplified naturally is done around the structural vibration computations described in previous sections; this determines the most efficient input for control (F_I^+). The shear layer grows spatially and is curved; these physical effects must be included in the stability analysis.

Stability theory and nonlinear parabolized stability equation (NPSE; Herbert 1997) simulations predict the most unstable or most effective frequencies for control by periodic perturbations. Recent experiments in artificially excited shear layers suggest that data are well correlated with nonlinear inviscid instability theory. On the basis of this and as a first step, spatial instability theory has been extended to a slowly divergent prescribed mean flow; in particular, a $\tanh(y)$ profile for two parallel streams of different speed that are allowed to mix. Preliminary results show that two-dimensional oscillations tend to increase in amplitude until a neutrally stable Strouhal number is attained. The results are in excellent agreement with the experiments of Wygnanski and Petersen (1987). Current calculations include the effects of the curvature characteristic of the free streamline leaving the leading edge of the lower surface and reattaching at the upper surface. To this end, the radius of curvature is varied parametrically to determine its effect on stability.

The next step is the application of the NPSE to the structural vibration computations described in previous sections. In recent years the NPSE have become a popular approach to analyzing streamwise growth of linear and nonlinear disturbances in slowly varying shear layers, jets, and boundary layers. The NPSE include nonparallel and nonlinear effects ignored by linear stability theory. Moreover, the NPSE have significantly less resource overhead associated with them compared with direct numerical simulations (Herbert 1997). Using a multiple-scales approach, each disturbance quantity is decomposed spectrally into a rapidly varying wave function and a slowly varying shape function in the spanwise and temporal directions. At finite ampli-

tude, a single mode of frequency ω_o and spanwise wavenumber β_o produces harmonics and steady mean-flow distortion. For 3-D applications, β_o is complex-valued. In the NPSE approach, the explicit streamwise second-derivative term is neglected based on physical arguments. Upon specification of initial and boundary conditions, the evolution of single or multiple modes can be tracked by marching downstream parabolically and simultaneously solving a coupled system for the relevant modes.

To date, the NPSE have been applied to a variety of 2-D and 3-D flow situations and are generally regarded as appropriate for convectively unstable flows (Haynes & Reed 1996; Reed et al. 1998). In particular, for a far more complicated wall-bounded shear layer, we validated the NPSE with the experiments of Reibert et al. (1996) and Saric et al. (1998) on an NLF(2)-0415 swept airfoil. A spanwise array of roughness elements is used near the airfoil leading edge to introduce 12-mm spanwise-periodic crossflow disturbances into the boundary layer. Fig. 18 shows the comparison of experimental velocity contours with the computed (NPSE) velocity contours. This comparison is made at 45% chord after this complicated flow has developed from the same initial conditions at 5% chord. It is clear that NPSE does an excellent job of capturing all of the details of the crossflow vortex structure with very little computational expense.

2 Conclusions

We have demonstrated that the Hurley flap is a good candidate for a flow-control experiment. It offers a new concept for high-speed wings, in particular, delta wings that may improve their low-speed characteristics and increase their lift. It may have other applications that require rapid flow deflections. For example: since the flow at the inlet of a stealthy airplane is required to turn while being decelerated, an interior flap of the Hurley type may prove effective over a wide range of angles of incidence and inlet flow conditions. The wing on a tilt rotor airplane of the "Osprey" type generates huge drag during a vertical climb and hover. A Hurley flap with active flow control may turn the flow efficiently thus reducing the drag penalty in this flight regime and increasing the useful load of the airplane. Applications to UAVs susceptible to gust and loll conditions are also straightforward.

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Sweep=60°, $\alpha=30^\circ$, $Re=4 \times 10^5$

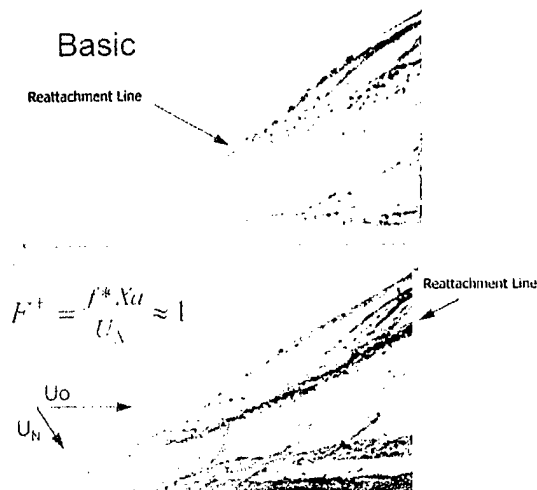


Figure 1. Flow control over delta wing.

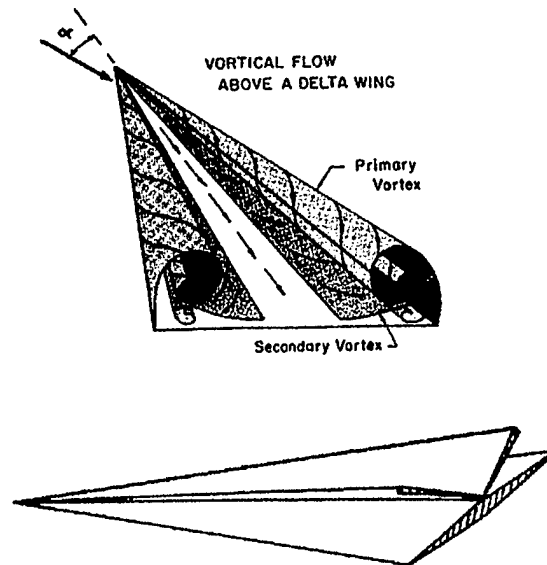


Figure 2. Vortical flow and Hurley flap as delta wing.

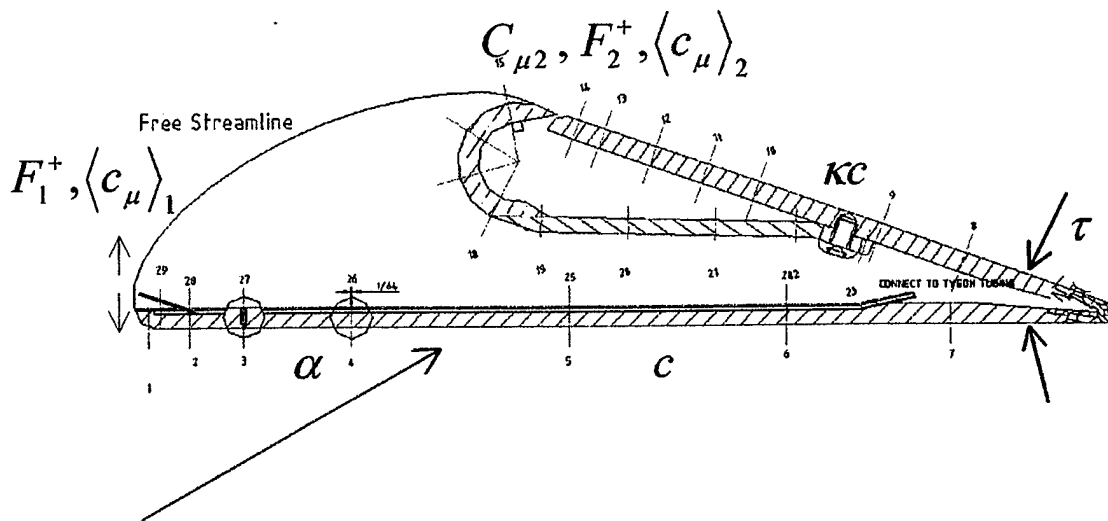


Figure 3. Significant parameters. Geometrical parameters are c , τ , κ , α , and slot location. Flow parameters are F_1^+ , $\langle c_\mu \rangle_1$, $C_{\mu 2}$, F_2^+ , $\langle c_\mu \rangle_2$, and Re .

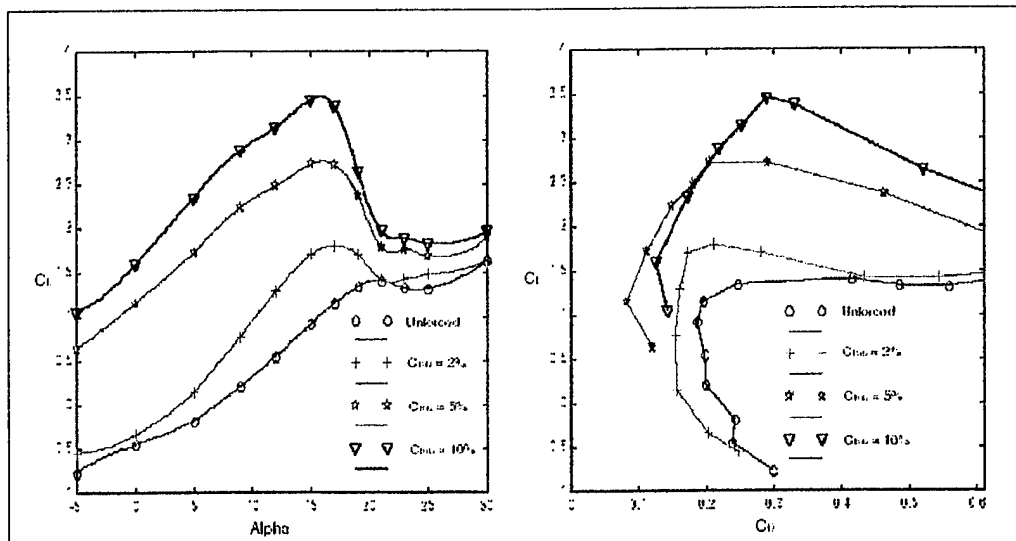
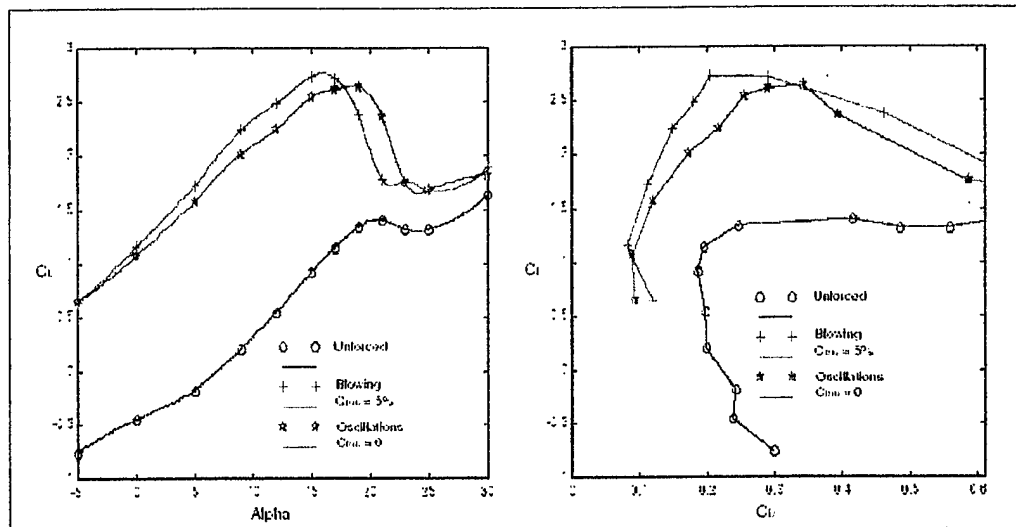
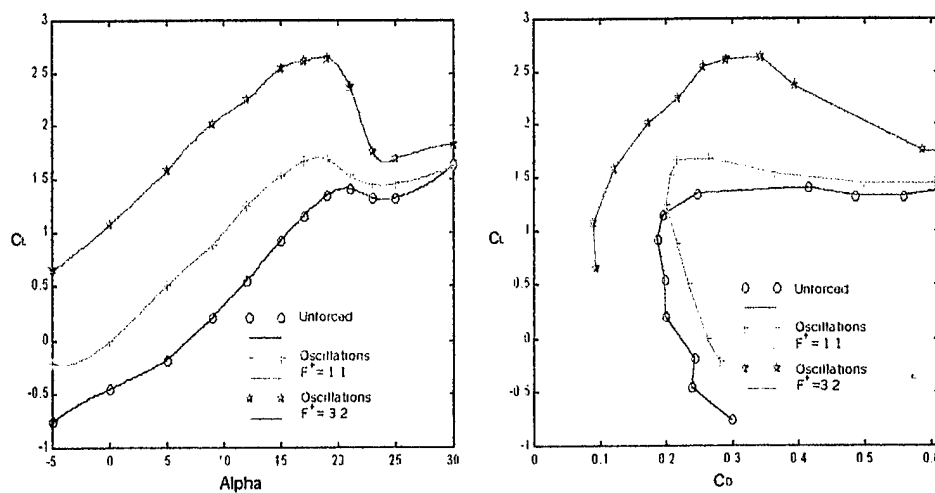
Figure 4. The effect of steady blowing. $Re = 150,000$.Figure 5. Influence of steady blowing and combined forcing. $Re = 150,000$. $\langle C_{mu} \rangle = 0.5\%$. $F^+ = 3.2$.

Figure 6. The Effect of Frequency on Lift and Drag

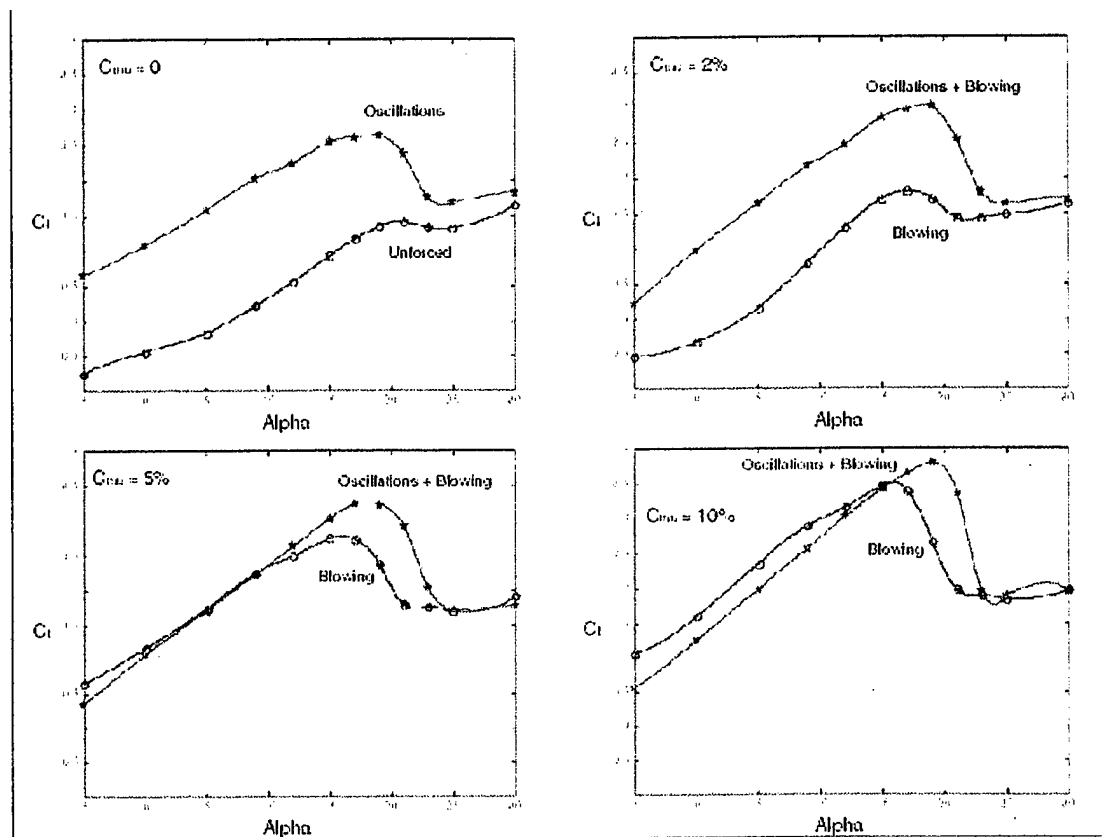


Figure 7. Comparison between blowing and combined forcing. $Re = 150,000$. $\langle C_{\mu} \rangle = 0.5\%$. $F^+ = 3.2$.

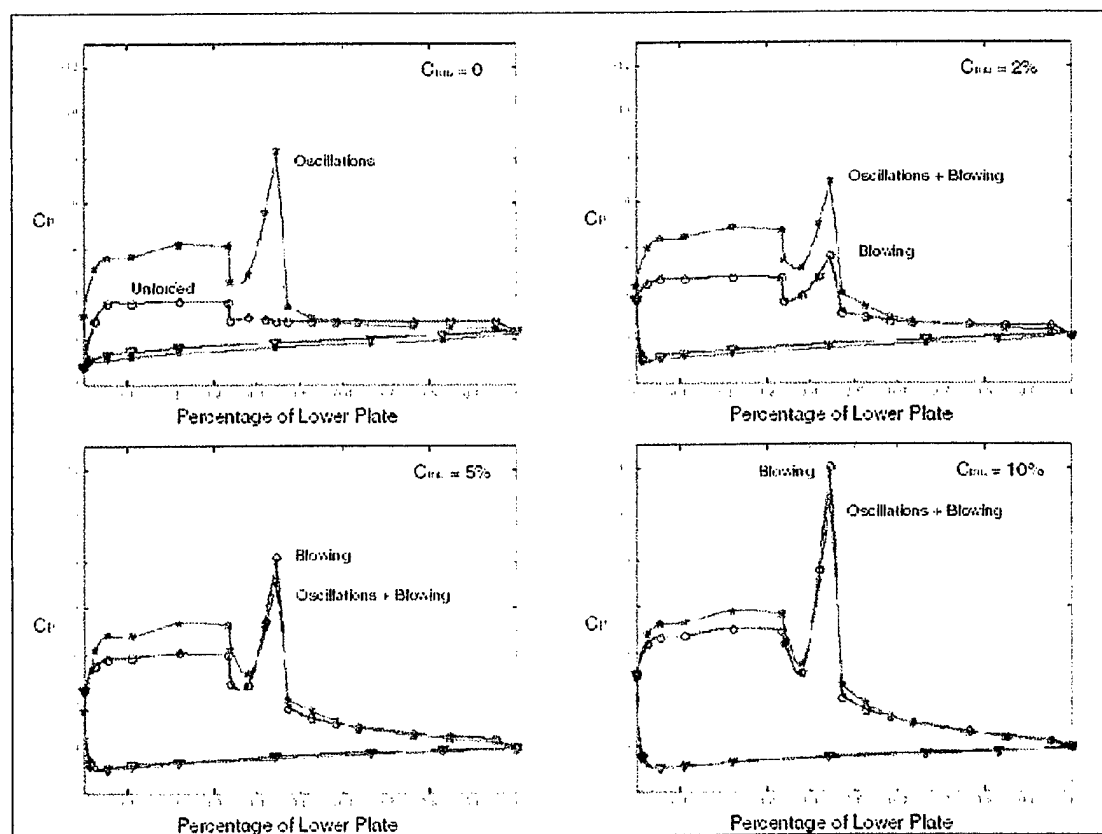


Figure 8. Pressure distribution at $\alpha = 17^\circ$. $Re = 150,000$. $\langle C_{\mu} \rangle = 0.5\%$. $F^+ = 3.2$.



Figure 9a. Flow visualization with oscillating blowing. $\langle C_{mu} \rangle = 0.5\%$. $\alpha = 15^\circ$.



Figure 9b. Flow visualization of steady blowing. $C_{mu} = 7\%$, $\alpha = 15^\circ$.

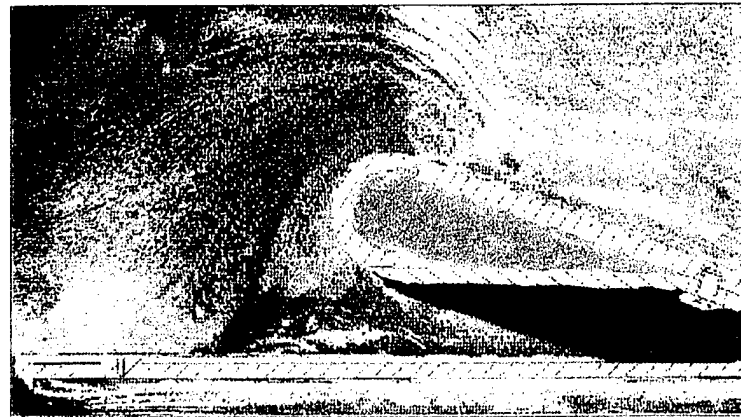


Figure 10a. Flow visualization of oscillating blowing. $\langle C_{mu} \rangle = 0.5\%$. $\alpha = 0^\circ$.

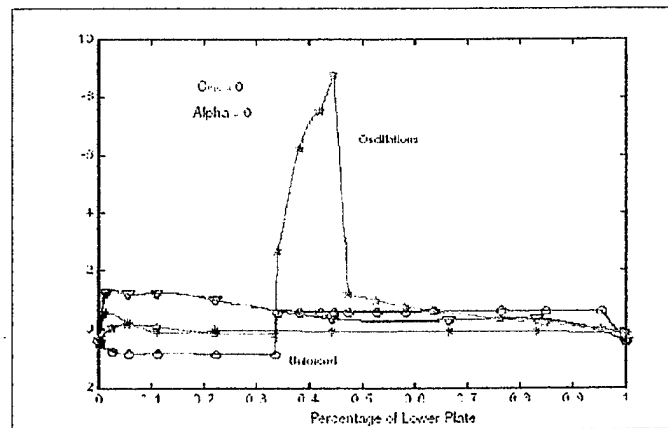


Figure 10b. Pressure distribution at $\alpha = 0^\circ$.

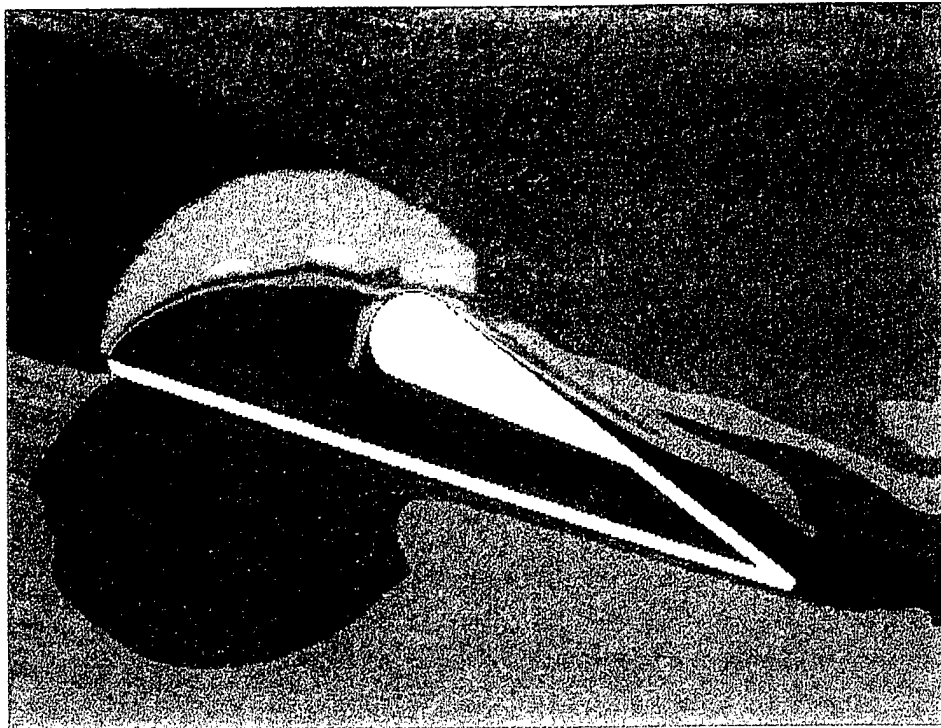


Figure 11a. Finite element LES calculation showing the instability of the free streamline.

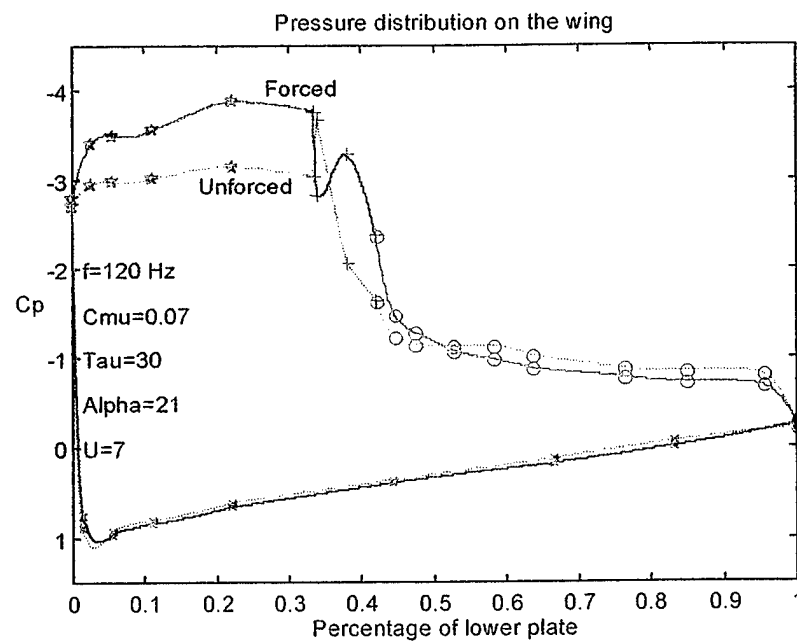


Figure 11b. Pressure distribution with and without the vibrating Flaperon.

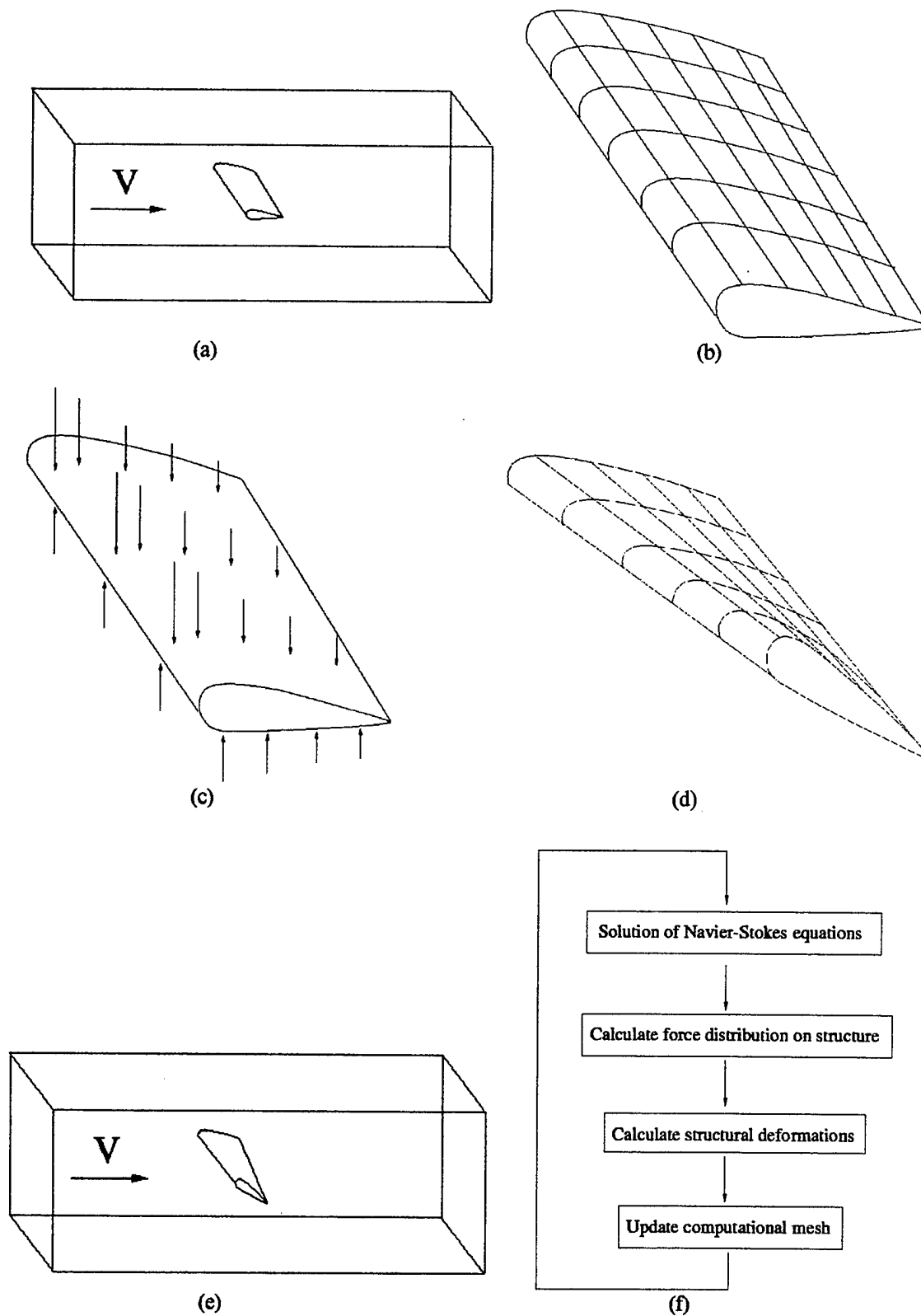


Figure 12: (a) Typical computational domain for fluid flow; (b) typical mesh for structural analysis; (c) transfer of pressure forces from fluid analysis to structural mesh; (d) deformation due to pressure forces; (e) updated computational domain for fluid flow; (f) the sequential fluid-structure interaction algorithm

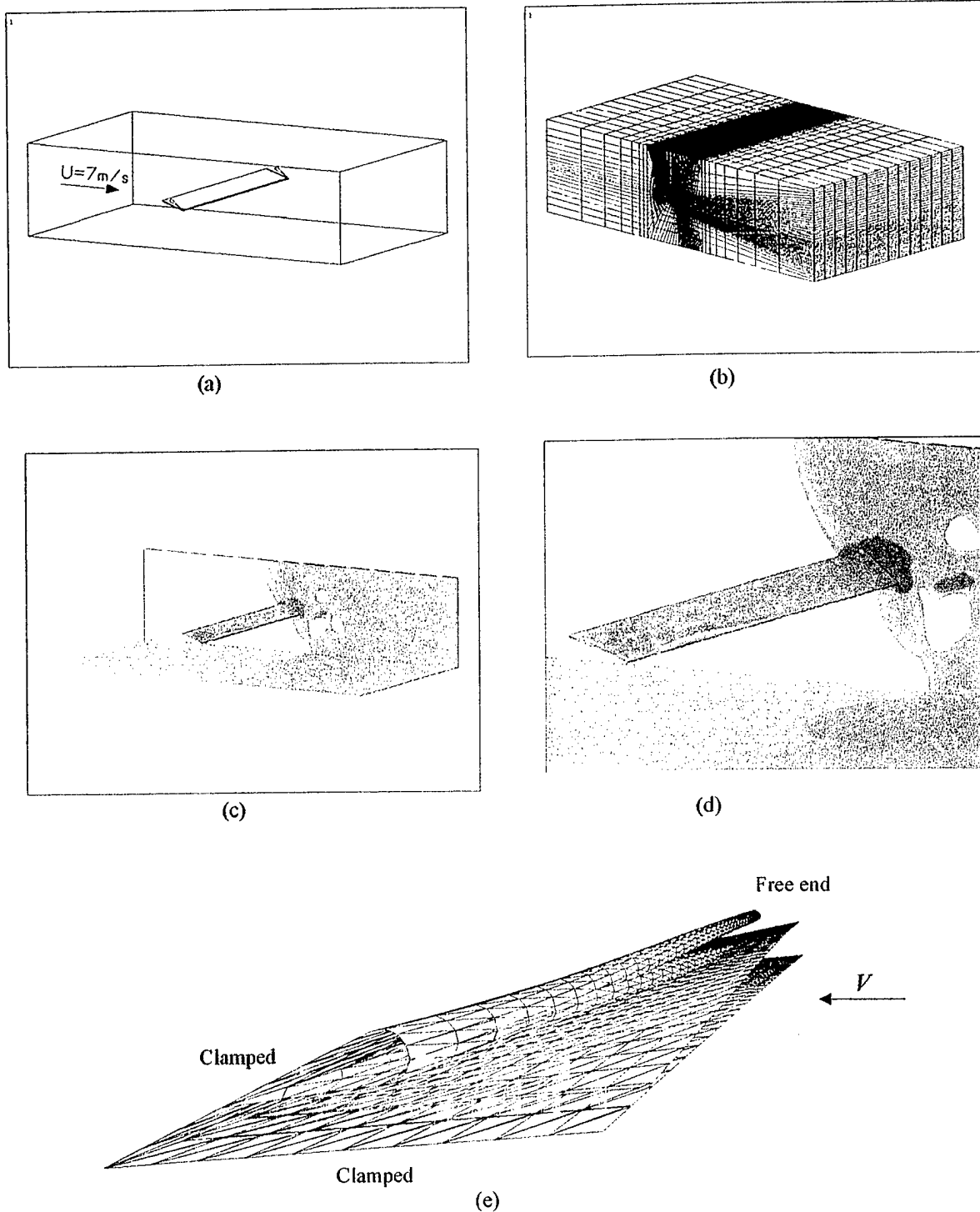


Figure 13. Analysis of Hurley wing, aspect ratio = 8: (a) computational domain, (b) fluid mesh (50,000 nodes), (c) pressure contours ($Re=1100$), (d) close-up view of pressure contours, and (e) deformation caused by fluid flow.

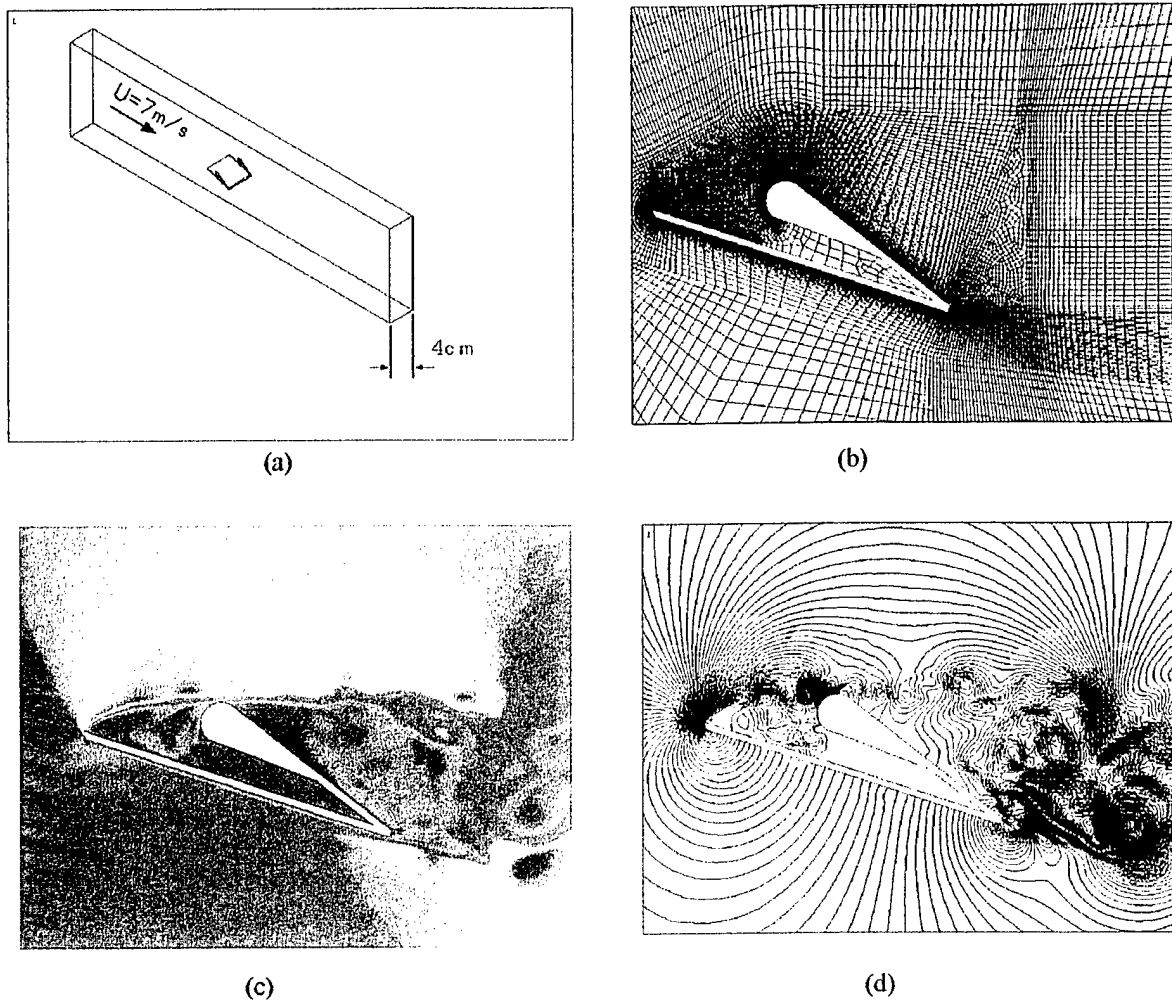


Figure 14. LES calculation at $Re = 120,000$: (a) computational domain (not to scale); (b) close-up view of a slice of the LES fluid mesh (200,000 nodes); (c) velocity contours from unsteady, 3-D LES, $C_L = 1.2$, $C_D = 0.28$ (Exp: $C_L = 1.2$, $C_D = 0.27$); and (d) pressure contours.

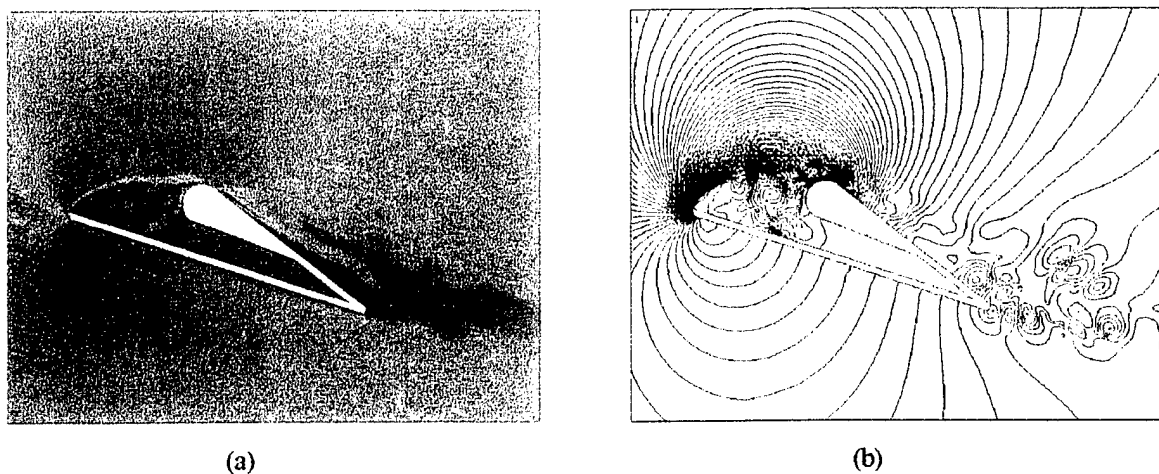


Figure 15. LES calculation at $Re = 120,000$, blowing slot with $C_{\mu} = 0.09$: (a) velocity contours from unsteady 3-D LES, $C_L = 2.5$ (Exp: 2.5); and (b) pressure contours.

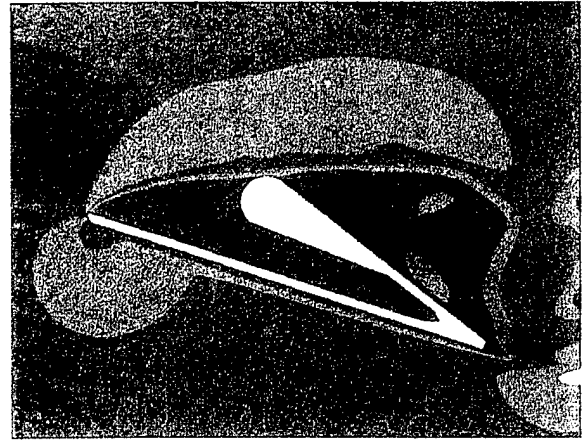
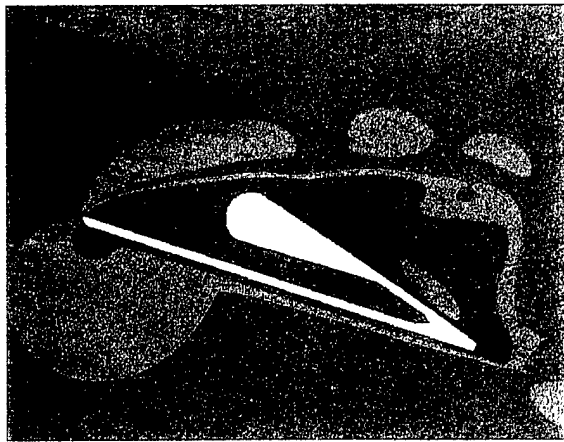
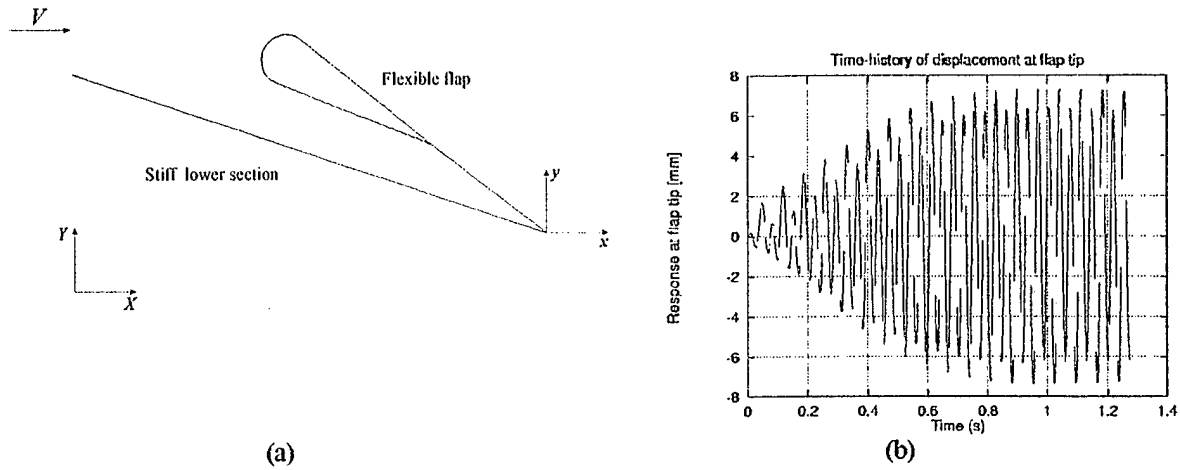


Figure 16. Fluid-structure interaction of Hurley wing with flexible flap: (a) idealized model of Hurley airfoil, (b) time-history of vertical displacement of upper flap tip, (c) velocity contours at bottom position, and (d) velocity contours at top position.

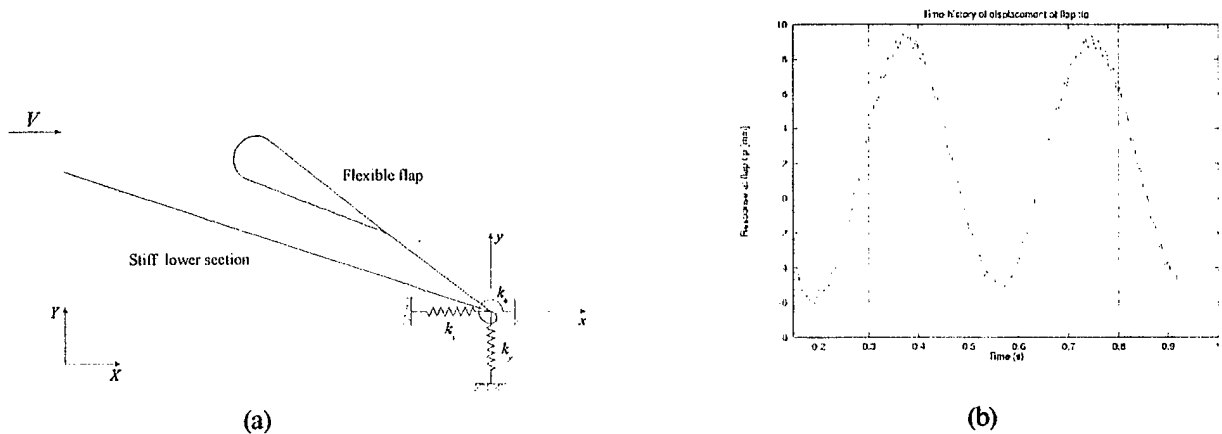


Figure 17. Fluid-structure interaction of Hurley wing attached to springs: (a) idealized model of the Hurley airfoil and (b) time-history of vertical displacement of upper flap tip.

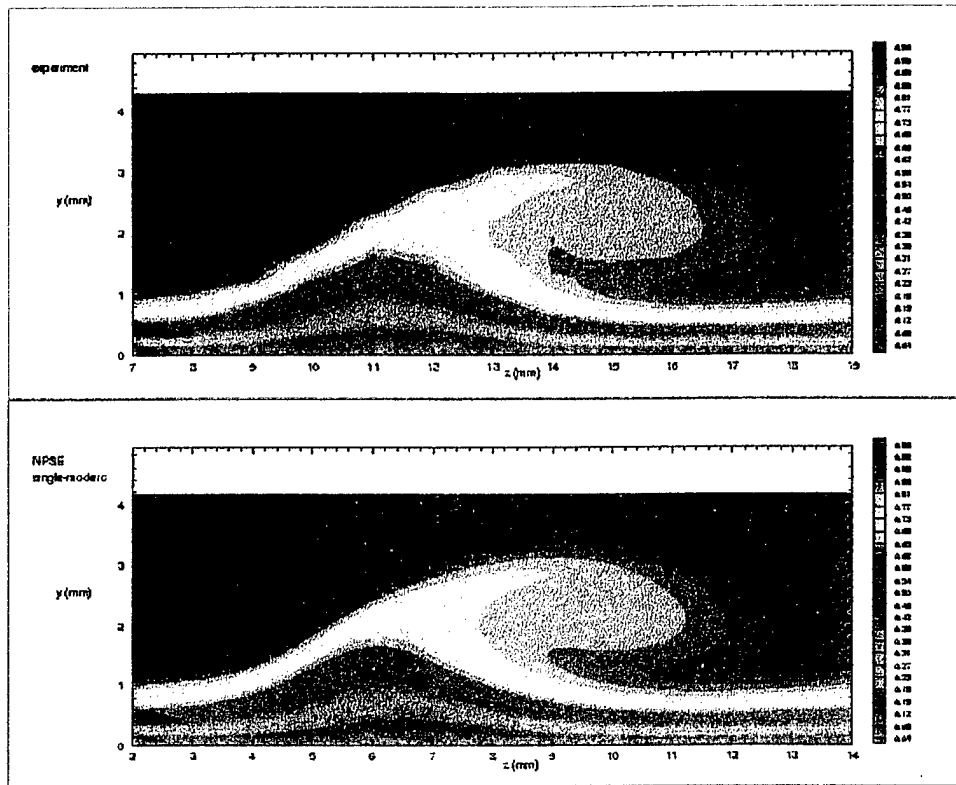


Figure 18. Velocity contours for stationary crossflow vortices at $Re_c = 2.4$ million. Upper plot is experimental data. Lower plot is nonlinear PSE calculation.

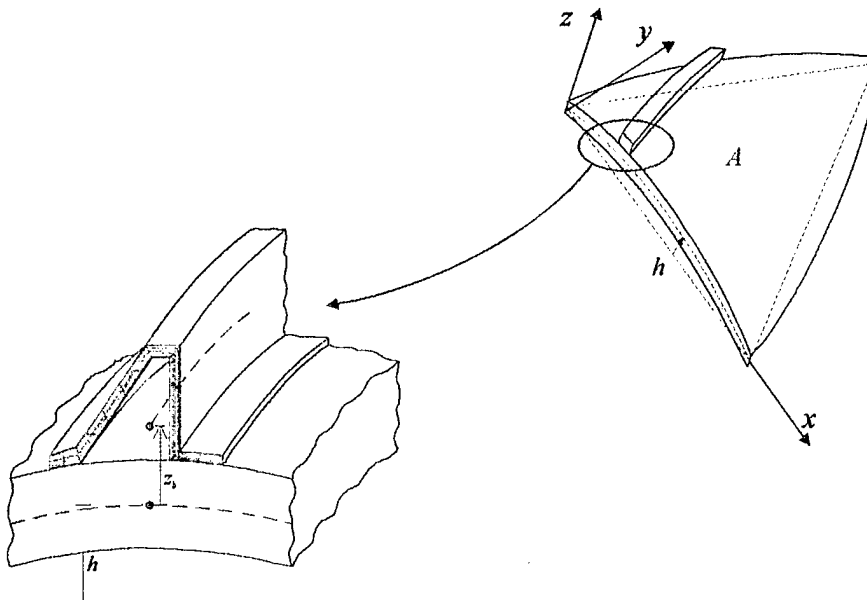


Figure 19. Close-up view of the triangular shallow stiffened shell element