

GAS GENERATOR ACTUATOR ARRAYS FOR FLIGHT CONTROL OF SPINNING BODY PROJECTILES

Brian A. English², Priya Gadiraju³, Christopher S. Rinehart², Ari Glezer², Mark G. Allen^{1,3}

Schools of Electrical and Computer¹, Mechanical², and Chemical³ Engineering

Georgia Institute of Technology, Atlanta, GA, USA 30332

ABSTRACT

This paper presents batch-fabrication lamination approaches for the realization of large arrays of high-power, short-duration gas generator actuators (GGAs), and system implementation approaches for integration of these GGAs into a 40-mm diameter gun-launched projectile for projectile flight control and course correction. The GGAs are tested to determine the impulse delivered per GGA and the time required delivering the impulse. The arrays of GGAs are connected with control electronics and integrated into a 40-mm diameter projectile. The final result is a flight control system for a small-scale projectile; fabrication and characterization of the actuator component of this system will be presented here.

1. INTRODUCTION

The GGAs are conceptually similar to previous microrockets [1] or micropyros [2, 3] in that solid fuel is combusted within a combustion chamber and exhausted through a nozzle. Solid fuel actuators are useful in applications that require disposable actuators to deliver large specific impulses with simplified fabrication. Unlike previous approaches designed for microsatellite applications, GGAs are designed to operate in atmospheric conditions and can take advantage of flow-control techniques in which the generated gas can interact with the embedding flow surrounding the moving projectile to produce larger-scale forces than those available from the GGA itself [4]. The platform for testing these devices is a gun-launched projectile with a 60 Hz spin rate. Large g-loads, on the order of 10g, are common at launch with gun-fired munitions, so robust

materials that are insensitive to shock are laminated to form the GGA arrays. Further complicating the application is the relatively high spin rate of these projectiles, requiring rapid (millisecond-scale) force rise and fall times, rendering many previous approaches inapplicable. This rapid force profile is achieved using appropriately-designed hot-wire igniters that create a larger combustion front, reshaping the force profile of the actuator, thereby generating larger forces in shorter times. Finally, the ignition and control system must be self-contained within the projectile. This requires that the ignition power be matched to the on-board power supply. The devices presented have the following characteristics

- Simplified combustor fabrication
- Robust materials, 10g launch
- Array of single-shot, disposable actuators
- Low voltage ignition, 8 VDC
- Rapid force rise/fall times, less than four millisecond total burn time

2. FABRICATION

GGA arrays were fabricated using laser micromachining and lamination techniques, **Figure 1**. The materials used to fabricate the GGAs were 250-micron thick Mylar with 10-micron thick pressure sensitive adhesive (PSA) bonded to both sides of the Mylar film, and the igniters were patterned from 15-micron thick Copper-Nickel film. These materials are rapidly patterned using laser machining, and they are robust enough to withstand the high g-load launch. The process for fabricating the combustors is as follows:

Report Documentation Page				Form Approved OMB No. 0704-0188	
Public reporting burden for the collection of information is estimated to average 1 hour per response, including the time for reviewing instructions, searching existing data sources, gathering and maintaining the data needed, and completing and reviewing the collection of information. Send comments regarding this burden estimate or any other aspect of this collection of information, including suggestions for reducing this burden, to Washington Headquarters Services, Directorate for Information Operations and Reports, 1215 Jefferson Davis Highway, Suite 1204, Arlington VA 22202-4302. Respondents should be aware that notwithstanding any other provision of law, no person shall be subject to a penalty for failing to comply with a collection of information if it does not display a currently valid OMB control number.					
1. REPORT DATE 2006		2. REPORT TYPE		3. DATES COVERED 00-00-2006 to 00-00-2006	
4. TITLE AND SUBTITLE Gas Generator Actuator Arrays for Flight Control of Spinning Body Projectiles				5a. CONTRACT NUMBER	
				5b. GRANT NUMBER	
				5c. PROGRAM ELEMENT NUMBER	
6. AUTHOR(S)				5d. PROJECT NUMBER	
				5e. TASK NUMBER	
				5f. WORK UNIT NUMBER	
7. PERFORMING ORGANIZATION NAME(S) AND ADDRESS(ES) Georgia Institute of Technology, School of Electrical and Computer Engineering, Atlanta, GA, 30332				8. PERFORMING ORGANIZATION REPORT NUMBER	
9. SPONSORING/MONITORING AGENCY NAME(S) AND ADDRESS(ES)				10. SPONSOR/MONITOR'S ACRONYM(S)	
				11. SPONSOR/MONITOR'S REPORT NUMBER(S)	
12. DISTRIBUTION/AVAILABILITY STATEMENT Approved for public release; distribution unlimited					
13. SUPPLEMENTARY NOTES The original document contains color images.					
14. ABSTRACT					
15. SUBJECT TERMS					
16. SECURITY CLASSIFICATION OF:			17. LIMITATION OF ABSTRACT	18. NUMBER OF PAGES 5	19a. NAME OF RESPONSIBLE PERSON
a. REPORT unclassified	b. ABSTRACT unclassified	c. THIS PAGE unclassified			

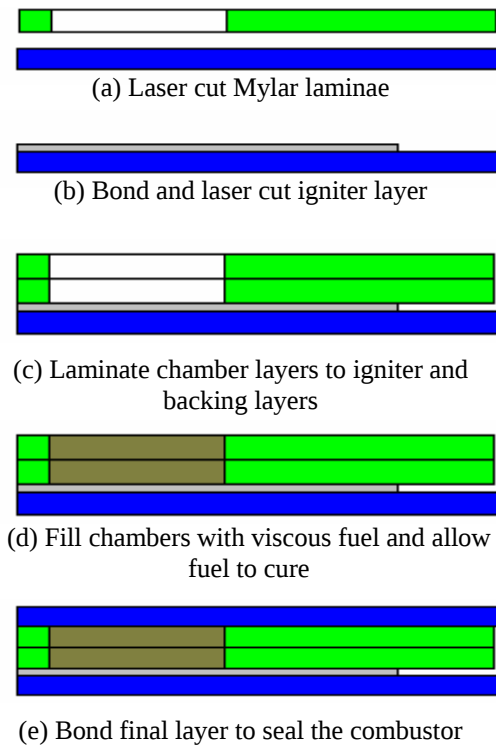
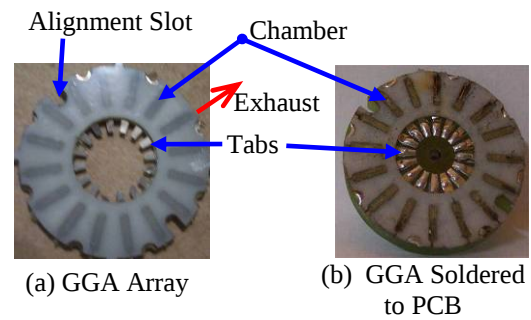


Figure 1. GGA Fabrication

- Pattern layers of Mylar using a CO₂ laser to form hollow combustion chambers and solid backing layers.
- Laminate 15-micron layer of Copper-Nickel film to a patterned layer of Mylar, align assembly to CNC controlled stage, and pattern Copper-Nickel foil with an IR-laser to form an igniter through the combustion chamber. The igniters are 500 x 250 x 15 microns and have a resistance of 1 Ohm. The igniters are patterned with small tabs that make the GGA array a surface mount component.
- Bond laminae with a hydraulic press at room temperature to form partially sealed combustion chambers. The combustion chambers are 3.3 mm³, and the nozzle throat is 600 microns x 150 microns. **Figure 2(c)** shows a view of a panel of GGA arrays at the completion of this step.
- Fill combustion chambers with viscous fuel, and then cure, or harden, the fuel.
- After the fuel has hardened, bond the final to seal the combustion chamber.



(c) 128 x 128 mm panel of GGA arrays Without fuel and final backing layer

Figure 2. Fabrication Results for 33 mm diameter radial array of Gas Generator Actuators

Figure 2 shows two views of the GGA assembly. **2(c)** is a panel of GGA arrays before the chambers have been filled with fuel. When fabrication is complete, individual GGA arrays are released by cutting the interconnecting plastic strips. This yields the GGA array seen in **2(a)**. The array is composed of 16 combustors aligned radially to a 30-mm diameter disk. The volume of each combustor is 3.3 mm³. The GGA array is aligned to a PCB, and the loose tabs are soldered to contact pads on the PCB as seen in **2(b)**. The top layer in **2(b)** was removed to show the cured fuel in the combustion chamber.

3. COMBUSTOR PERFORMANCE

The GGA arrays are designed to integrate with an axisymmetric, bluff body with a spin rate of 60 Hz. The combustors must deliver an impulse within a 4 ms rotation of the body in order to be delivering force in the desired direction. This action steers the body towards the opposite direction. The ignition delay must be well controlled so the combustor fires in the control direction, making actuator reproducibility an

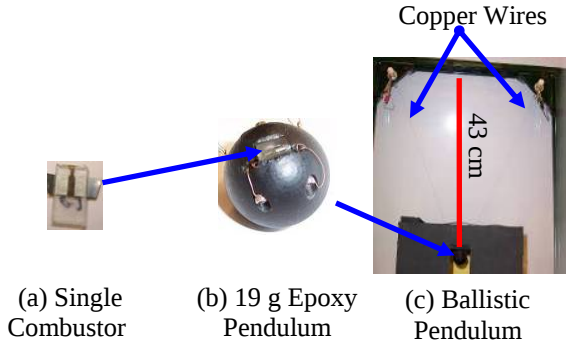


Figure 3. Ballistic Pendulum Components

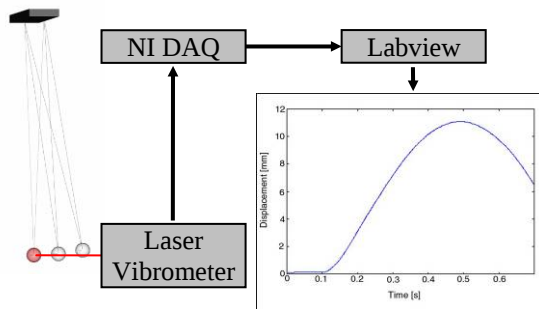


Figure 4. Ballistic Pendulum Setup

important manufacturing constraint. Therefore, the total impulse delivered, the impulse duration, and the ignition delay are characterized in order to control the application of forces to the spinning body. Combustors are characterized using a fixed force balance and a ballistic pendulum. Finally, the GGA arrays are integrated into a windtunnel model and the jet was observed interacting with the flow field.

3.1 Ballistic Pendulum

Total GGA Impulse and Specific Impulse of the propellant are determined using a ballistic pendulum, **Figure 3**. Single combustor chips, **3(a)**, were loaded into an epoxy ball, **3(b)**, and this ball was suspended by two copper wires to create a pendulum arm length of 43 cm, **3(c)**. Electrical power is conducted through the copper wires and then conducted through a mechanical contact to the combustor chip. A laser vibrometer records the position of the ballistic pendulum, and data is digitally recorded. **Figure 4** shows the measurement system and a sample vibrometer output. The initial velocity, initial displacement, and maximum displacement of the

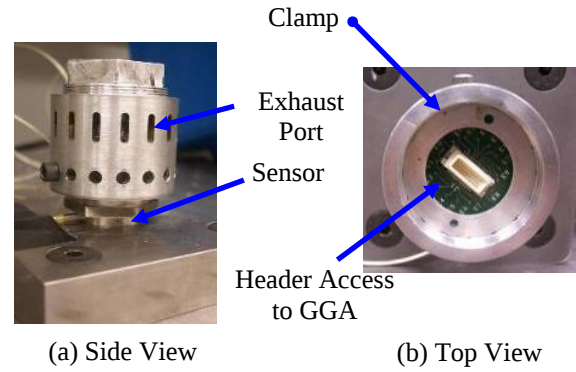


Figure 5. Force Stand Setup

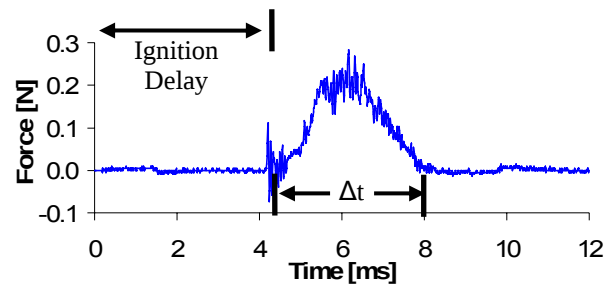


Figure 6. Force Response

ballistic pendulum are used to calculate the total impulse delivered from the combustor. Testing has shown that the 3.3 mm^3 devices filled with polymeric fuel produce impulses of $1 \pm 0.22 \text{ N}\cdot\text{ms}$. Combustor mass is measured before and after firing to determine the specific impulse of the fuel that combusted. The mass loss was measured to be $2.2 \text{ mg} \pm 0.7 \text{ mg}$. The average specific impulse for polymeric fuel mixture tested was 49 s.

3.2 Force Balance

Ignition delay and impulse duration are determined using a stationary force balance, **Figure 5**. A Kistler piezoelectric force sensor is used to record force applied by the GGAs. The GGAs are clamped into the rotating turret, and product gases escape through machined exhaust ports. Electronic access to each combustor is passed through a central header. The combustor turret is linked to the force sensor through a damped ring to reduce the amount of high-frequency oscillations recorded by the force sensor. **Figure 6** shows a typical force response.

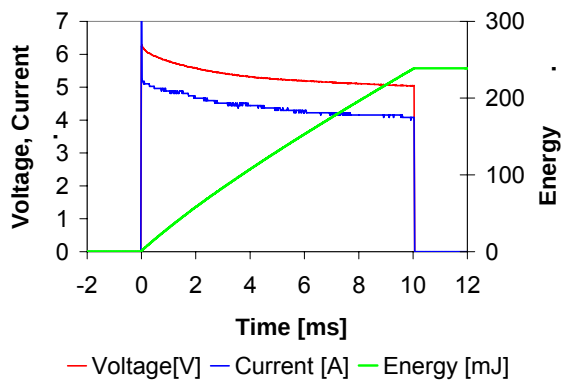


Figure 7. Electrical Input for Ignition

A peak force of 250 mN is recorded 5.8 ms after ignition, and the duration of the force is 3.3 ms. The ignition delay is 4.2 ms when determined to be the start of the combustion event. For the 3.3 mm³ combustor presented, the ignition delay is 4.3 $\pm$ 0.3 ms and the combustion duration is 3.7 $\pm$ 1 ms. Integrating the force curve yields impulses of 0.6 $\pm$.2 N-ms. These values are smaller than impulse values recorded using the ballistic pendulum because of the damping ring between the combustor turret and the force sensor. The force results show that the combustor impulse can be sufficiently directed as the projectile spins.

3.3 Electrical Operation

The power supply for the GGA arrays consists of two 200mAh Li-ion batteries in parallel with a hybrid-tantalum, 5 mF capacitor. Drive circuitry activates a power mosfet to control the pulse duration to the igniter. **Figure 7** shows the ignition energy delivered to the igniter. The battery voltage with no-load is approximately 8VDC. When the mosfet is activated, 6.2 VDC and 5.2 A is delivered to the igniter. Voltage and current decrease to 5VDC and 4 A as the capacitor discharges and the igniter resistance increases. The total energy delivered is 250 mJ.

3.4 Windtunnel Operation

Finally, system operation is verified by imaging the jet exhaust in a wind tunnel **Figure 8** illustrates successful wind tunnel testing with the GGA exhaust jet penetrating into a 35 m/s

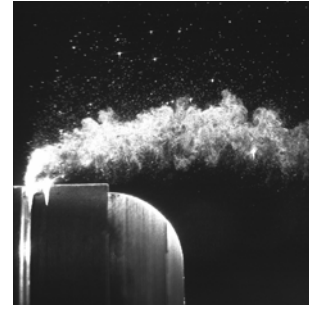


Figure 8. Gas expulsion in cross flow

crossflow. The jet penetrates into the flow field before being swept downstream. This type of exhaust locally separates the boundary layer from the body, increasing the area of the trailing wake.

4. CONCLUSION

The GGA arrays demonstrate a simple fabrication method to produce robust arrays of solid fuel actuators. These actuators are capable of delivering large impulses within millisecond timescales, and the repeatable ignition delay allows for the implementation of these actuators onto a moving, or spinning body. The power supply has been reduced to two batteries, a capacitor, and control electronics which would also facilitate the application of GGA arrays into independent packages. Windtunnel imagery shows significant flow field penetration for a 35 m/s crossflow, suggesting that the combustor forces are on the order of convective flow field forces for this free-stream velocity.

ACKNOWLEDGEMENT

This work is supported in part by DARPA as part of the Micro-Adaptive Flow Control program.

1. Teasdale, D., *Solid Propellant Microrockets*, in *Electrical Engineering and Computer Sciences*. 2000, University of California at Berkley: Berkley. p. 46.
2. Rossi, C., D. Esteve, and C. Mingues, *Pyrotechnic actuator: a new generation of Si integrated actuator*. *Sensors and Actuators A*, 1999. 74: p. 211-215.

3. Rossi, C., et al., Design, fabrication and modeling of solid propellant microrocket-application to micropropulsion. *Sensors and Actuators A*, 2002. **99**: p. 125-133.
4. Crittenden, T.M., *Fluid Actuators for High Speed Flow Control*, in *Mechanical Engineering*. 2003, Georgia Institute of Technology: Atlanta, GA. p. 190.