



COMPARATIVE ANALYSIS
OF A HIGH BYPASS TURBOFAN
USING A PULSED DETONATION COMBUSTOR

THESIS

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AFIT/GAE/ENY/07-M02

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Abstract

It has been proposed that the implementation of a pulsed detonation combustor in a high-bypass turbofan engine would result in an engine that is both more efficient and more reliable. The validity of the performance claims are evaluated based on a comparison between the baseline and hybrid turbofans as modeled in the Numerical Propulsion System Simulation (NPSS). The engine cycle of the baseline high-bypass turbofan is evaluated and compared using both the Aircraft Engine Design System (AEDsys) and NPSS programs. The baseline engine agreed to within 1% on the net thrust calculation and specific fuel consumption between the two programs. Differences are traceable to the variation in specific heat models and to different methods of calculating temperature across the turbine and compressor components. The hybrid pulsed detonation engine model shares a common architecture with the baseline turbofan model. Inlet mass flow and core mass flow are maintained, but the combustor of the baseline engine is replaced with a pulsed detonation combustor for the hybrid engine. The effect of detonation on the core air flow is calculated using a closed form solution of the Chapman-Jouguet Mach number with a total energy correction applied. Cycle time is calculated to provide a reasonable estimate of frequency for the user input geometry. Effects of sub-component design choices within the pulsed detonation combustor are evaluated using simple parametric studies. These studies are used to select an optimal architecture for the combustor. The effects of detonation are accounted for by applying pressure and temperature losses to the fluid exiting the combustor. A parametric study was performed to demonstrate what level of loss continues to yield a more efficient engine. Results show improvement in thrust specific fuel consumption with careful selection of combustor design parameters. There is a definite level of acceptable loss that if surpassed makes pulsed detonation combustion a good candidate for inclusion into a hybrid turbofan engine.

Acknowledgements

This thesis represents more than one man's work. Their names may not appear on the cover page, but it belongs as much to those who have helped as to me.

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My children have a debt of gratitude. You sacrificed time with your Dad, weekend trips to the zoo and museum, bedtime stories, and one-on-one time so I could do homework and a thesis. Only time will tell if the sacrifice you made in these early years will give you more of the father that you need.

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Ionio Q. Andrus

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List of Symbols

Symbol		Page
ρ	Density	4
p	Pressure	4
γ	Ratio of Specific Heats	4
q	Heat Flux	4
h°	Heat of Reaction	4
u_{CJ}	Chapman-Jouguet Detonation Wave Velocity	4
FAR	Fuel-to-Air Ratio	4
h_{pr}	Fuel Lower Heating Value	4
u	Fluid Velocity	5
\dot{m}	Mass Flow Rate	5
a	Speed of Sound in a Fluid	6
\tilde{q}	non-dimensional Heat Addition	8
C_P	Constant Pressure Specific Heat	8
f	Mass Fuel-to-Air Ratio	8
h_{pr}	Fuel Lower Heating Value	8
Ψ	Non-dimensional Temperature Ratio	8
M_{CJ}	Chapman-Jouguet Mach Number	8
S	Entropy	8
η_b	Burning Efficiency	8
π_b	Burner Pressure Ratio	8
η_N	nozzle efficiency	9
C_v	Nozzle Gross Thrust Coefficient	10
h_0	Enthalpy at Beginning of Cycle	10
q_{add}	Heat Flux Into System	10
F _n	Net Thrust	14

Symbol		Page
BPR	Bypass Ratio	14
T_{t4}	Total Temperature at Turbine Inlet	18
$\frac{\Delta P}{P}$	Unscaled Normalized Pressure Drop	18
CTOL	Low Pressure Spool Power Take-off Coefficient	19
CTOH	High Pressure Spool Power Take-off Coefficient	19
π_{dmax}	Maximum Inlet Pressure Ratio $\left(\frac{P_{t2}}{P_{t1}}\right)$	19
π_b	Burner Pressure Ratio $\left(\frac{P_{t4}}{P_{t3}}\right)$	19
π_n	Main Nozzle Inlet Pressure Ratio $\left(\frac{P_{t9}}{P_{t7}}\right)$	19
π_{nf}	Fan Nozzle Pressure Ratio $\left(\frac{P_{t19}}{P_{t17}}\right)$	19
η_c	Compressor Adiabatic Efficiency	20
η_{tb}	Turbine Adiabatic Efficiency	20
e_c	Compressor Polytropic Efficiency	20
e_{tb}	Turbine Polytropic Efficiency	20
\dot{W}	Work Rate or Power	20
P_r	Reduced Pressure	21
π_c	Compressor Total Pressure Ratio	21
h	Enthalpy	21
π_{tb}	Total Pressure Ratio across the Turbine	21
τ_{tb}	Total Temperature Ratio Across the Turbine	21
h_{tbexit}	Turbine Exit Enthalpy	21
P_t	Total Pressure	22
ϕ	Entropy Function	25
Δh	Difference in Enthalpy Between Two Models	29
pf	Purge Fraction	41
ff	Fill Fraction	41
A_{tube}	Detonation Tube Cross-Sectional Area	42
l_{tube}	Tube Length	42

Symbol		Page
$V_{purge\ air}$	Purge Air Volume Per Cycle	42
$V_{fuel-air\ mixture}$	Fuel-Air Mixture Volume per Cycle	42
m_{purge}	Purge Air Mass	42
$m_{fuel-air\ mixture}$	Fuel-Air Mass for Fill	42
ρ_t	Total Density	42
$m_{fill\ air}$	Mass of Pure Air Mixed with Fuel During Fill	43
m_{fuel}	Mass of Fuel Used for One Fill Cycle	43
$\dot{m}_{air\ total}$	Mass Flow Rate of Air Through Detonation Tubes	43
f	Filling or Cycle Frequency	43
\dot{m}_{vo}	Maximum Mass Flow Rate when Detonation Valve is Open	43
MFP	Mass Flow Parameter	43
τ_{vo}	Valve Open Time Fraction $\frac{t_{valve\ open}}{t_{cycle}}$	44
t_{cycle}	Cycle Time	44
DDT	Deflagration to Detonation Time	44
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F _n	Net Thrust	67
TSFC	Thrust Specific Fuel Consumption	67

List of Abbreviations

Abbreviation		Page
PDE	Pulsed Detonation Engine	1
PDC	Pulsed Detonation combustor	1
SFC	Specific Fuel Consumption	1
NPSS	Numerical Propulsion System Simulation	2
AEDsys	Aircraft Engine Design System	2
CJ	Chapman-Jouguet	4
ZND	Zeldovich, Von Neumann, and Döring	6
CVCCE	Constant Volume Combustion Cycle Engine	10
TIT	Turbine Inlet Temperature	14
FPR	Fan Pressure Ratio	14
OPR	Overall Pressure Ratio	14
PR	Pressure Ratio	16
LPC	Low Pressure Compressor	17
HPC	High Pressure Compressor	17
HPT	Low Pressure Turbine	17
LPT	High Pressure Turbine	17
LP	Low Pressure	17
HP	High Pressure	17
RPM	Revolutions Per Minute	18
PTO	Power Take Off	19
VSH	Variable Specific Heats	20
FAR	Mass Fuel-to-Air Ratio	29
iBPR	Internal Bypass Ratio	47
TSFC	Thrust-Specific Fuel Consumption	51
NIST	National Institute of Standards and Technology	60

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I. Introduction

A significant effort has been made in the past ten years to develop pulsed detonation engines (PDE) as a means of aircraft propulsion. Detonation combustion holds the promise of a more efficient engine due to the simultaneous pressure increase and combustion. Additionally, fewer moving parts mean lower maintenance costs in association with this engine.

Efficiency is the impetus for development, but PDEs still face several key technical hurdles. Noise is a significant issue for any aviation engine, and *detonation* creates more of it than previous aircraft engines.

It is hoped that a marriage of the PDE with traditional turbomachinery will result in an engine that is more efficient than turbine engines and quieter than a pure PDE. The engine configuration of interest to this work involves replacing the burner or high pressure spool of a turbofan with a pulsed detonation combustor (PDC). The studies of Petters and Felder [1] and Smith et al. [2] have indicated that specific fuel consumption (SFC) might be improved by 5 to 15% with the inclusion of a PDC. These claims are significant, and they require explanation through some sort of thermodynamic modeling.

1.1 Purpose

This thesis was generated in order to better understand how PDC technology will impact future engines. Specifically, the objective of this thesis was to generate a model for a PDC-turbofan hybrid engine, and to compare the hybrid model to a conventional (or “baseline”) turbofan engine. Pursuing the hybrid engine cycle only

makes sense if the comparison demonstrates an increase in propulsive efficiency. Thus, the goal of the comparison is to test the claims that a hybrid engine cycle is more efficient than the baseline technology.

1.2 Procedure

Two steps were taken to ensure results demonstrated only the change in combustion technology. First, the baseline high bypass turbofan with separate exhaust streams was modeled in two programs to corroborate the engine cycle solutions. Then a PDC hybrid turbofan was modeled and compared with the baseline engine.

1.2.1 Turbofan Comparison in NPSS and AEDsys. The baseline high bypass turbofan engine was modeled in the Numerical Propulsion System Simulation (NPSS) and in the Aircraft Engine Design system (AEDsys). The engine cycle calculation result comparison assured us that the engine configuration was identical between the two programs and acceptable to the propulsion community.

Each piece of software was selected for a different reason. In order to allow the models generated for this thesis to be useful to the sponsor (AFRL/PRTA), NPSS was mandated for the both the baseline and hybrid engines. NPSS was developed within the cooperative framework established between industry and NASA, and is becoming an industry standard. AEDsys was selected as the second piece of software because it is easy to use, available, and well documented. AEDsys was developed by Mattingly [3] for use in preliminary and academic situations. The comparison was not intended to demonstrate the accuracy of either code, nor to provide a qualitative judgment of either code. Instead, the intent was to ensure that the results generated by the model constructed in NPSS are reasonably correct.

1.2.2 Comparison of PDC Hybrid with Baseline Turbofan. Once the baseline engine was well-established the thermodynamic performance of a PDC was coded into an NPSS element based on the work of on the work of Heiser and Pratt [4] and

Dyer and Kaemming [5]. This NPSS element was then called by the NPSS hybrid engine model to perform cycle analysis calculations. This hybrid model required some minor configuration changes to the baseline engine's high pressure section in order to accommodate the new element.

An acceptable pulsed detonation configuration had to be identified before the baseline and hybrid engines could be compared. This configuration was determined through parametric studies which showed how various PDC parameters affected the overall engine performance. Once an acceptable PDC configuration was identified, a component impulse analysis was performed using NPSS for both the hybrid and baseline engines at design conditions. The effects of the unsteady and noisy flow generated by the pulsed detonation in the hybrid cycle performance was then simulated through the application of pressure and temperature losses to the flow exiting the combustor.

The results indicate that a hybrid engine can be more efficient than a conventional high-bypass turbofan engine. Care must be taken to prevent significant total pressure loss as the flow is transitioned to a steady state in order to see the improvement. Detonation tube fill time has a significant impact on the density of the engine - shorter tube fill times will result in fewer tubes being required to provide the same thrust.

II. Prior Work on Pulsed Detonation Combustors

A significant amount of work has been performed to model pulsed detonation combustors. This chapter reviews some of the detonation theories, detonation engine performance work, PDC models, and the results generated from those models. The models are central to the pulsed detonation process, which in turn is central to the hybrid engine cycle.

2.1 Thermodynamics of Pulsed Detonation Engines

The models seeking to mimic actual pulsed detonation are based on the underlying thermodynamics. Both Kuo [6] and Glassman [7, 221-265] have good explanations of two basic models of detonation summarized in sections 2.1.1 and 2.1.2.

2.1.1 Chapman Jouquet Detonation. Only a few brief points on Chapman-Jouquet (CJ) theory are given here since Kuo [6] provides a detailed description and derivation. CJ theory builds on the (Rankine-) Hugoniot relation defined by:

$$\frac{\gamma}{\gamma - 1} \left(\frac{p_2}{\rho_2} - \frac{p_1}{\rho_1} \right) - \frac{1}{2} (p_2 - p_1) \left(\frac{1}{\rho_1} + \frac{1}{\rho_2} \right) = q \quad (2.1)$$

where ρ is density, p is pressure, and γ . When exit pressure p_2 is plotted as a function of specific volume $\frac{1}{\rho_2}$ for a constant heat flux (q) value, the result is the Hugoniot curve. Recall that q is defined as

$$q = h_1^\circ - h_2^\circ \quad (2.2)$$

where h° is the heat of reaction. So we can control the offset of the Hugoniot curve (and CJ wave velocity u_{CJ}) by changing the q value through fuel-to-air ratio (FAR) or fuel lower heating value (h_{pr}).

The Hugoniot curve is traditionally broken into five regions, as shown in Figure 2.1. Consider the Rayleigh line equation in region V where both density (ρ) and

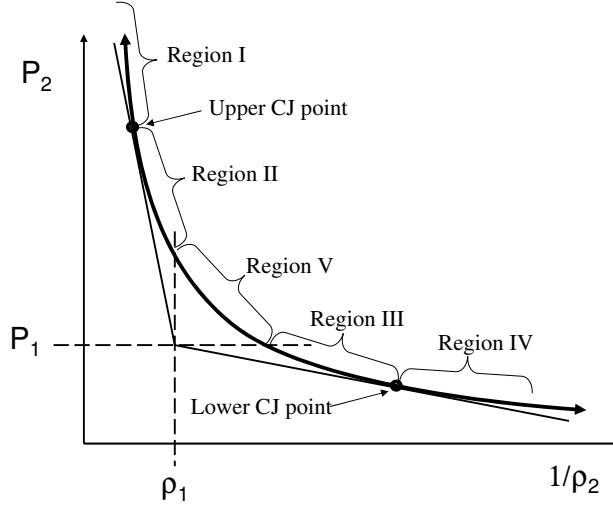


Figure 2.1: Diagram of Hugoniot curve. Combustion is generally only observed in region III and at the upper Chapman-Jouguet points.

pressure (p) at station 2 are greater than at station 1:

$$\rho_1^2 u_1^2 = \frac{p_2 - p_1}{\frac{1}{\rho_1} - \frac{1}{\rho_2}} = \dot{m}^2 \quad (2.3)$$

For these conditions, equation 2.3 implies an imaginary solution for the fluid velocity (u) or mass flow rate (\dot{m}). Region I is rarely observed, requiring an overdriven shock. Region II is likewise rarely observed, requiring a fast-reacting mixture. Region IV will require that the gas be accelerated transonically through the deflagration wave and is therefore defined as forbidden. The lower CJ point bordering region IV is also not seen experimentally and is therefore excluded. This leaves only region III - referred to as weak-deflagration - and the upper Chapman-Jouguet point. Since detonation is the region of interest to this work we will ignore deflagration and focus on the upper CJ point.

The Chapman-Jouguet points are located where the following equality holds true:

$$\frac{p_2 - p_1}{\frac{1}{\rho_2} - \frac{1}{\rho_1}} = -\gamma p_2 \quad (2.4)$$

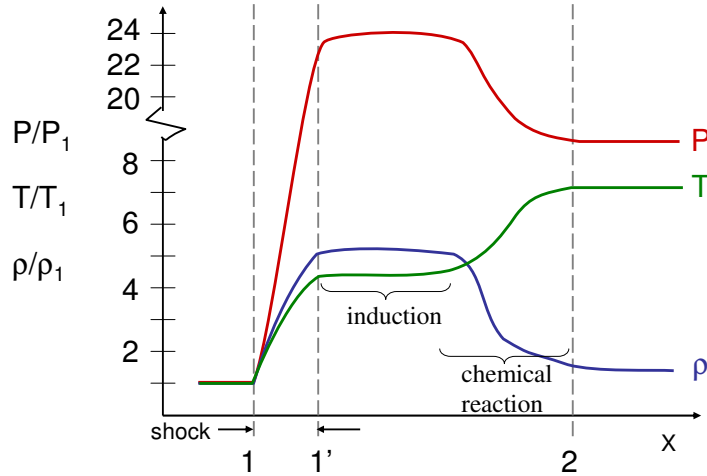


Figure 2.2: Diagram of the 1-dimensional detonation model used for ZND detonation theory.

This equality can be combined with equation 2.3 to yield equation 2.5.

$$u_2^2 = \frac{\gamma p_2}{\rho_2} = a_2^2 \text{ or } |u_2| = a_2 \quad (2.5)$$

where a is the speed of sound in the fluid. This implies that at the CJ points, $M_2 = 1$. The upper CJ point corresponds to the minimum detonation wave speed. Since commonly observed detonation corresponds to the upper CJ point, we seek to obtain the corresponding velocity. There are several methods of doing this that are iterative in nature, but a closed-form solution has also been obtained and will be discussed in section 2.1.3.

2.1.2 The Zeldovich-Von Neumann- Döring and Multi-Dimensional Detonation Models. Zeldovich, Von Neumann, and Döring (ZND) independently made the same assumption that detonation flow is one dimensional. The result is the ZND model of detonation, extending the Chapman-Jouguet theory of detonation. Their model treats the detonation as a shock wave propagating into the quiescent mixture that pre-heats the reactants to a level that allows the ensuing combustive reaction to travel at the speed of the shock. This is easily seen in Figure 2.2. As the unreacted

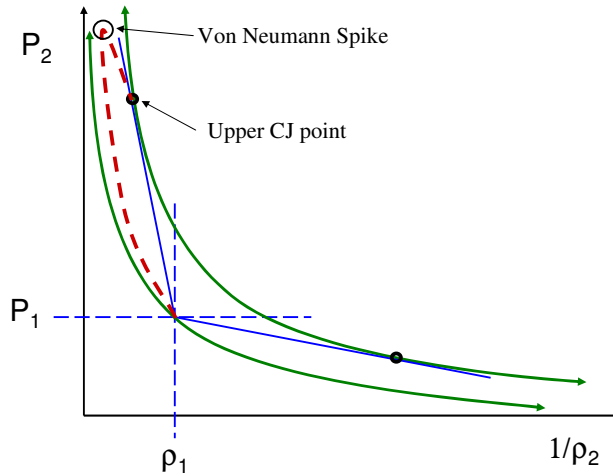


Figure 2.3: As the detonation initiates, it drives the fluid properties along the dashed path, causing a pressure spike first seen by von Neumann.

gas ($q = 0$) passes through the shock, the pressure rise drives it along the Hugoniot curve to the point described in Figure 2.3 as the von Neumann spike. It is at this point that combustion begins to occur, releasing heat and moving the fluid to a new Hugoniot curve with a different value of q . Height of the spike depends on the rate of chemical kinetics, with faster chemical interactions resulting in a smaller spike. If the spike is not high enough, or non-existent, the detonation may not have enough power to sustain itself.

These models demonstrate the underlying effects of detonation, but the reality is always more complex. Detonation occurs in three dimensions, and is a complex interaction where pressure, temperature, and density vary in all directions as shown in Kuo [6] and Strehlow [8]. Detonations are cellular in nature, so the geometry of the detonation tube can have a dramatic impact on the ability to reliably detonate reacting mixtures. Indeed, for a cylindrical tube, the detonation wave prefers to travel in a spiral whose pitch for the leading transverse wave is defined by the first helical acoustic mode.

2.1.3 *Thermodynamic Performance Models.* Heiser and Pratt [4] presented a simplified model for thermodynamic performance calculation of a PDE. Their method is based on solving for the Chapman-Jouguet Mach number of the detonation wave flowing into the quiescent mixture. The CJ Mach number is then used to calculate entropy and pressure rise across the detonation volume. With these properties, they can then calculate all fluid properties of the fluid exiting the PDE. The closed form algebraic solutions developed by Shapiro and others [8–10] are summarized here.

First, a non-dimensional heat addition term is calculated

$$\tilde{q} \equiv \frac{q_{supplied}}{C_p T_0} = \frac{f h_{pr}}{C_p T_0} \quad (2.6)$$

where \tilde{q} is the non-dimensional heat addition, C_p , f is the mass fuel-air ratio, and h_{pr} is the lower heating value of the fuel. Having calculated \tilde{q} it is possible to calculate the Chapman-Jouguet Mach number using equation 4 from Heiser and Pratt [4]:

$$M_{CJ}^2 = (\gamma + 1) \left(\frac{\tilde{q}}{\Psi} \right) + 1 + \sqrt{\left[(\gamma + 1) \left(\frac{\tilde{q}}{\Psi} \right) + 1 \right]^2 - 1} \quad (2.7)$$

where Ψ equal $\frac{T_3}{T_0}$, the ratio of the compressed static temperature to the free-stream static temperature. Having calculated the Chapman-Jouguet Mach number (M_{CJ}) it is possible to calculate the entropy S generated using equation 5 from Heiser and Pratt [4]:

$$\frac{s_4 - s_3}{C_p} = -\ln \left[M_{CJ}^2 \left(\frac{\gamma + 1}{1 + \gamma M_{CJ}^2} \right)^{\frac{\gamma+1}{\gamma}} \right] \quad (2.8)$$

For a real PDE cycle, the non-dimensional heat addition term, \tilde{q} , is multiplied by a burning efficiency (η_b) and a pressure ratio (π_b) is calculated across the tubes using equation 18 from Heiser and Pratt [4]

$$\frac{p_4}{p_0} = \frac{1 + \gamma M_{CJ}^2}{\gamma + 1} \frac{p_3}{p_0} \geq 1 \quad (2.9)$$

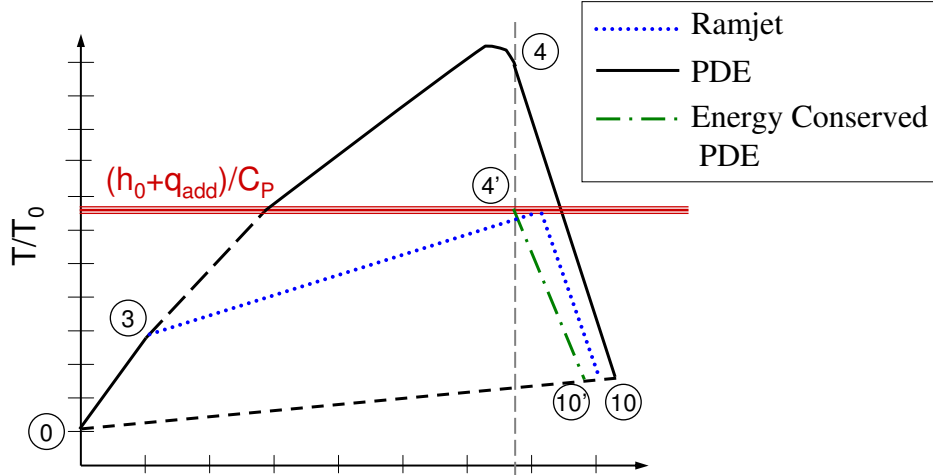


Figure 2.4: The T-S diagram shown in Dyer and Kaemming paper [5, fig 10] with the proposed correction to the PDE thermodynamic cycle. Conservation of energy will prevent incorrect calculation of entropy as the fluid exits the engine.

With both the entropy and pressure of the fluid post-combustion known it is possible to solve for all the other fluid properties through look-up tables or linear approximations.

Harris et al. [11] faulted the method of Heiser and Pratt [4] as overly optimistic because it neglects the fill portion of the cycle. Neglecting the fill portion does not invalidate the solutions obtained for pressure and entropy, but it does change the time averaged value of the properties. There is, however, a more significant drawback to using the cycle solution proposed by Heiser and Pratt [4]: energy is not conserved.

Dyer and Kaemming [5] proposed a different thermodynamic performance estimation and compared their results with ramjet thermodynamic performance. Their work is very similar to the work of Heiser and Pratt [4], but differs in how it accesses the fluid properties at station 4 (detonation tube exit). Figure 2.4 shows a temperature-entropy diagram of three cycles: a ramjet cycle, the PDE cycle as proposed by Heiser and Pratt, and the energy conserved PDE cycle. They realized that using a nozzle efficiency (η_N) accesses the kinetic energy at station 4, an energy level that is not truly available since the velocity will be “paid back” when gasses expand

back to static conditions. Instead, they use the nozzle gross thrust coefficient (C_v) that utilizes the entropy at tube exit and ambient pressure - not the energy. The result is that less entropy is calculated between generation from tube exit (station 4) and system exit (station 10). They propose that an accurate representation of the available energy is to use the known CJ entropy as calculated by Heiser and Pratt [4] coupled with the known system enthalpy calculated as $(h_0 + q_{add})$ where h_0 is the enthalpy at the beginning of the cycle and q_{add} is the heat flux into the system.

2.2 Pulsed Detonation Combustor Models

Embedding pulsed detonation within a turbine engine has been examined by groups from NASA [1, 12, 13], G.E. Global Research [14–20], the U.S. Air Force [4, 21], and Indiana University and Liberty Works [2, 22]. The results of this work have generated hope that a marriage between pulsed detonation and turbomachinery will result in a more efficient engine that is simultaneously mechanically simpler. Work initiated at NASA has been moved to industry through the Constant Volume Combustion Cycle Engine program (CVCCE)

2.2.1 Pulsed Detonation Combustor Performance Maps. Work done at General Electric Global Research [17] under the CVCCE program implemented a CFD code to achieve a converged pulse detonation cycle that described fluid elements and cycle timing. CFD results were ported into a performance map as described by Paxson [13]. This approach uses the CFD cycle results for inlet temperature, fraction of time that the flow contains a reacting mixture, and mixture stoichiometry to calculate PDC performance. The map returns total enthalpy and total pressure ratios given the three input parameters of 1) fill fraction 2) purge fraction and 3) non-dimensional heat of reaction (q_0). There were some minor modifications to Paxson’s [13, 23] formulae, but the process remained unchanged. This method was nearly selected for implementation in this work because of the ability to base a performance map on experimental data, but time limitations drove the work in a different direction.

The CFD simulation was a 1-D MacCormack predictor-corrector time marching algorithm, second order accurate in time and space. A 1-D simulation does not capture all the effects of inlet valves and combustor exhaust, however, it does provide information about change in fluid properties at combustor exit. Also, cycle time calculations that provided accurate detonation frequency and mass flow rates. Significantly, a parametric study to demonstrate the effect of constant specific heats was performed and included in the report by Tangirala et al. [17]. It demonstrated that maintaining a constant specific heat through a PDC analysis results in a flawed prediction of total pressure ratios and cycle times.

Additionally, the stated geometry of the detonation tube used for the PDC was not realistic: its length was only 3 diameters long. This is not long enough to allow for a detonation cycle to occur. Although not stated, the assumption is that since the model is one dimensional only the length of the tube, and total cross-sectional area are important to find a solution.

2.2.2 Exhaust into Turbines. One major hurdle to the implementation of a PDC into hybrid engines is the interaction between the cyclic pulsation of the detonations and the flow through the turbines. In order to maintain the efficiency of the baseline engine, a pulsed engine will require efficient power extraction from the combusted fluid. A good understanding of the pulsed detonation exhaust is required if current turbine technology is to be used.

A significant body of work exists for PDE applications, and results have converged to agreement over the past five years or so. The data that has converged to agreement for unconfined venting of PDE exhaust. A PDC, on the other hand, vents into a volume confined by the detonation tube exhaust plane, splitter plates, and the stator/turbine assembly. Schauer et al [21] noted that when venting into a turbine, blowdown times of a single PDE tube increased from 3 ms to more than 10 ms, often extending into the fill portion of the next cycle. This delay subsequently impacts the

fill negatively. Recognizing this fundamental difference, Rasheed et al. [14] performed some research to better understand this process.

Under the NASA CVCCE program, GE Global Research [16,18–20] performed a series of tests on an eight-tube PDC exhausting into a turbine taken from a locomotive turbocharger. The detonation tubes used in this experiment were un-valved; flow entered the inlet plenum perpendicular to the detonation tubes, and tube cooling air was flowed through a separate inlet. These conditions differ from those that were modeled for our work, but the results gained from this effort provide valuable insight into how a PDC will work in conjunction with a turbine.

In an effort to better understand how a pulsed detonation engine affects the flow through the turbine, the G.E. Global Research group [14] also performed a series of CFD and experimental studies examining how the detonation waves interacted with a 2-dimensional cascade of stator blades. Their work shows that there is significant fluctuation in mass flow rate, pressure, and temperature across the turbine during the pulse detonation cycle. The fluctuations induced by the detonation are passed through the turbine and will affect the other rotor stages as well.

2.2.3 Converting Pulsed Flow into Steady State Flow. The results reported by Schauer et al. [21] and G.E. Global Research [19,20] indicate that current turbine technology will not perform as expected unless the pulsed nature of the PDC is somehow balanced to produce steady-state or near steady-state flow into the turbines.

The machinery required between the pulsed flow generated by a PDC and the steady flow desired for optimal turbine performance has not yet been identified. It may involve ejectors, capacitive tanks, diffusers or a combination of these items combined with new ideas. It seems reasonable to assume that whatever the machinery, some pressure and enthalpy loss to the fluids that traverse them. The level of losses is not known at this time, but certainly there will be a definite efficiency required before the hybrid technology can present an improvement over a baseline high bypass turbofan engine.

We are left with the broader question: what level of losses can be incurred in the transition to steady state before the benefit gained by pulsed detonation is lost?

III. Baseline and Hybrid Models

Models of the baseline and the hybrid high bypass turbofan were generated with great care to ensure consistency. The baseline model was evaluated using both AEDsys and NPSS. This secured a good understanding of the high bypass turbofan and provided a solid foundation for comparing it to the hybrid engine. Once the baseline model was defined, the hybrid model implementing the PDC could be completed .

3.1 Baseline High Bypass Turbofan

The modeled engine was based on public information of the TF-39-GE-1C engine currently used on the C-5 Galaxy to certify that it was both relevant to the Air Force and easily verifiable. Table 3.1 shows a comparison of available information on the TF-39-GE-1C and the engine model generated for this exercise. The engine model was created without implementation of a requirements analysis and remains notional at this point. Since the component efficiencies were unknown, they were selected from Mattingly’s table 4.4 [3, 107], using a level 4 technology. Level 4 technology corresponds not to the engine technology currently used, but to the technology level projected ten to twenty years in the future. The level 4 technology also projects a higher turbine inlet temperature (TIT) than the engines currently found in service.

Table 3.1: Fundamental parameters for baseline engine.

Engine	TF39-GE-1C	Notional Baseline
Aircraft	C-5	N/A
Fn (lb)	40805	41500
Weight (lb)	7186	N/A
Length (in)	100	N/A
Max Diam (in)	203	N/A
BPR	8	8
FPR	1.56	1.56
OPR	26	26
TSFC (1/h)	0.315	0.325
Airflow (lbm/s)	1549	1500
TIT (F)	2350	2830

3.1.1 Baseline Engine Configuration. The engine model coded in NPSS is constrained by the fixed architecture of the high bypass split stream turbofan engine described by Mattingly [3, 569-587]. Indeed, the engine was first modeled using AEDsys with the input parameters given in Tables 3.2, 3.3, and 3.4. This high bypass turbofan engine is described in the most fundamental level by figure 3.1. The engine employs a separate exhaust stream for the bypass. A mixer for combining the fluid exiting the burner and one of the bleeds is implemented between stations 4 and 4.1. A similar mixer is implemented between stations 4.4 and 4.5. Fixed conical nozzle assumptions are used, and the fan is the outer portion of the first stage on the low-pressure compressor disk. Figure 3.1 shows the bypass and core airflows splitting at the fan exit, but the NPSS architecture required the split at the fan entry for correct implementation of separate fan and low-pressure compressor maps. The model file that defines the NPSS engine can be found in Appendix A. A component diagram of the engine is shown in Figure 3.2

Table 3.2: AEDsys reference design input variables for the baseline turbofan engine and thermodynamic package selection.

Input Variables	Value	Units
Gas Model		
Constant, Modified, Variable Specific Heat	MSH	
Reference Engine Design		
Flight Conditions		
Mach Number	0	
Altitude	0	ft
Temperature	518.67	R
Pressure	14.696	psia
Size		
Mass Flow Rate	1500	lbm/s
Design Limits		
total temperature leaving combustor	2900	R
Design Variables		
Compressor Pressure Ratio (Pt3/Pt2)	26	
LPC PR (Pt2.5/Pt2)	1.56	
Fan PR (Pt13/Pt2)	1.56	
Bypass ratio	8	

Table 3.3: AEDsys thermodynamic inputs for the modified specific heat option.

Input Variables	Value	Units
Fuel/Gas properties, Component Efficiency		
Fuel/Gas Properties		
Fuel Heating Value	18400	BTU/lbm
C_P compressor	0.2415	BTU/(lbm R)
gamma c	1.3986	
C_P turbine	0.295	BTU/(lbm R)
gamma turbine	1.2957	
Bleed air/ Coolant air		
Bleed Air	1	%
Coolant Air	5	%
Coolant Air 2	5	%
Power Takeoff		
C_{TOL}	0.00	
C_{TOH}	0.0005330	
LP Spool Mechanical Efficiency	0.99	
HP Spool Mechanical Efficiency	0.99	
Component total Pressure Ratio		
π diffuser max (Pt2/Pt1)	0.995	
π burner (Pt4/Pt3)	0.96	
π nozzle (Pt9/Pt7)	0.985	
π fan nozzle	0.98	
Polytropic Efficiencies		
Fan	0.89	
LPC	0.89	
HPC	0.90	
HPT	0.89	
LPT	0.89	
Component Efficiencies		
Burner	0.995	
Mechanical shaft LP spool	0.99	
Mechanical shaft HP spool	0.99	

Table 3.4: AEDsys engine control inputs.

Input Variables	Value	Units
Controls / Install Model/ # Engines		
Engine Controls		
Max Temperature at Station 4 (T_{t4})	3200	R
Max Compressor Pressure Ration	32	
Max Pressure at station 3	650	psia
Max Temperature at station 3	1860	R
Max % Ref RPM - LP Spool	110	
Max % Ref RPM - HP Spool	110	

3.2 Baseline Model Similarity Between AEDsys and NPSS

Although the baseline turbofan engine shares a common architecture in both programs, the two softwares require different methods of characterizing the components. Great care had to be taken to ascertain that the configurations not only had the same form, but also shared the same definition of component performance. Inevitably, differences in each program's solution process will produce some variation in the results. This section describes the greatest contributor to solution differences.

Design parameters were carefully ported into the NPSS model from the AEDsys program to eliminate differences due to component performance inputs. NPSS handles efficiencies of several types differently than AEDsys, so a translation had to be made in order to confirm models were similar. For some components, efficiencies input in AEDsys had to be translated to a normalized pressure drop term $\frac{\Delta P}{P}$ for NPSS. For other components, the calculated values of efficiency output from AEDsys had to be copied and input into the NPSS model. Table 3.5 shows the translation between the two sets of inputs. Also note that mechanical efficiencies from AEDsys translate to fractional losses in NPSS. The most significant item in this table is that NPSS requires adiabatic efficiency input whereas AEDsys requires polytropic efficiency for the compressors and turbines as an input.

Engine control inputs are not included in Table 3.5 because there is not a simple element that will impose restrictions on the engine. Instead, the control limitations

Table 3.5: AEDsys inputs differ slightly from NPSS inputs. This table should provide a method to correlate the inputs required by each program.

AEDsys Input	NPSS Element	Name in Appendix	Variable	Translation
Gas Model	Thermo Package	N\A	GasTbl\janaf\etc.	NPSS thermo translates to AEDsys' VSH
Mach number	FlightConditions	AMBIENT	MN	Same
Altitude	FlightConditions	AMBIENT	alt	Same
Temperature	FlightConditions	AMBIENT	ZTt and ZTs	must set both
Pressure	FlightConditions	AMBIENT	ZPt and ZPs	must set both
Mass Flow Rate	FlightConditions	AMBIENT	W	Same
T_{t4}	Burner	BRN36	TtCombOut	set switchBurn=TEMPERATURE
Compressor PR	Compressor	HPC25	S_map.PRdes	CPR = HPC _{PR} *LPC _{PR}
LPC PR	Compressor	LPC20	S_map.PRdes	Same
Fan PR	Compressor	Fan21	S_map.PRdes	Same
Bypass Ratio	Splitter	SPLIT	BPR	Same
Fuel Heating Value	FuelStart	FUEL32	LHV	Depends on thermodynamic package.
Bleed Air (% \dot{m})	Bleed	BLD3	port name.fracW	Divide AEDsys value by 100
Coolant Air 1 (% \dot{m})	Bleed	BLD3	port name.fracW	Divide AEDsys value by 100
Coolant Air 2 (% \dot{m})	Bleed	BLD3	port name.fracW	Divide AEDsys value by 100
CTOL	Shaft	LPShf	HPX	HPX = CTO* h_0 * \dot{m}_0/η_m
CTOH	Shaft	HPShf	HPX	(may also use AEDsys engine test output)
LP Spool PTO Efficiency η_m	Shaft	LPShf	fracLoss	Included in HPX calc
HP Spool PTO Efficiency η_m	Shaft	HPShf	fracLoss	if different than mechanical efficiency
$\pi_{dmax} \left(\frac{P_{t2}}{P_{t1}} \right)$	Inlet	INLET	eRamBase	Same
$\pi_b \left(\frac{P_{t4}}{P_{t3}} \right)$	Burner	BRN36	dPqPBase	$\frac{\Delta P}{P} = 1.0 - \pi_b$
$\pi_n \left(\frac{P_{t9}}{P_{t7}} \right)$	Nozzle	Noz8	dPqP	$\frac{\Delta P}{P} = 1.0 - \pi_b$
$\pi_{nf} \left(\frac{P_{t19}}{P_{t17}} \right)$	Nozzle	Noz18	dPqP	$\frac{\Delta P}{P} = 1.0 - \pi_b$
Fan polytropic efficiency	Compressor	Fan21	S_map.effDes	(Use AEDsys output isentropic efficiency)
LPC polytropic efficiency	Compressor	LPC20	S_map.effDes	(Use AEDsys output isentropic efficiency)
HPC polytropic efficiency	Compressor	HPC25	S_map.effDes	(Use AEDsys output isentropic efficiency)
HPT polytropic efficiency	Compressor	HPT41	S_map.effDes	(Use AEDsys output isentropic efficiency)
LPT polytropic efficiency	Compressor	LPT45	S_map.effDes	(Use AEDsys output isentropic efficiency)
η_b	Burner	BRN36	effBase	Same
η_m LP Spool	Shaft	LPShf	fracLoss	fracLoss = 1.0 - η_m
η_m HP Spool	Shaft	HPShf	fracLoss	fracLoss = 1.0 - η_m

are imposed through the use of independent and dependent variable declarations. These declared variables are then added to the NPSS solver. More information can be found in the NPSS user guide [24, sec 6.8].

With two different types of input for efficiency, a significant certification effort ensured that the turbines and compressors had the same performance. The polytropic and adiabatic efficiencies could not be matched simultaneously between the engines because the two programs work the progression from the input efficiency to the calculated efficiency in reverse order from each other.

Recognizing the need to relate the polytropic and isentropic (or adiabatic) efficiencies, a review of a fundamental turbomachinery text was performed. At a simplified level, Oates [25, 214 & 222] and Wilson and Korakianitis [26] recorded the formulae for computing the adiabatic efficiencies (η_c and η_{tb}) given the polytropic efficiencies (e_c and e_{tb}):

$$\eta_c = \frac{\frac{\pi_c^{\frac{\gamma_c-1}{\gamma_c}} - 1}{\pi_c^{e_c \gamma_c} - 1}}{\frac{\pi_c^{\frac{\gamma_c-1}{\gamma_c}} - 1}} = \frac{\dot{W}_{in\ ideal}}{\dot{W}_{in\ actual}} \quad (3.1)$$

$$\eta_{tb} = \frac{1 - \pi_{tb}^{\frac{e_{tb}(\gamma_{tb}-1)}{\gamma_{tb}}}}{1 - \pi_{tb}^{\frac{\gamma_{tb}-1}{\gamma_{tb}}}} = \frac{1 - \tau_{tb}}{1 - \tau_{tb}^{\frac{1}{e_{tb}}}} = \frac{\dot{W}_{out\ actual}}{\dot{W}_{out\ ideal}} \quad (3.2)$$

where \dot{W} is the power put into or taken out of the component. However, this relationship is observed in AEDsys only for the constant and modified specific heat thermodynamic cycles. Using this relationship to calculate the adiabatic efficiency does not account for the variation of specific heat calculated with the AEDsys variable specific heats (VSH) option or NPSS thermodynamic packages. The formulae for variable specific heats become more complicated since the variation of temperature across the components requires a complete thermal properties solution at exit.

With the goal of understanding the difference between AEDsys and NPSS, an examination of the methods employed by these programs was made. Mattingly [3, 106] chose to use the polytropic efficiency as input for his VSH because it is a measure of

technology. The polytropic efficiency is used to obtain the reduced pressure (P_r) at compressor exit:

$$P_{r\ exit} = P_{r\ inlet} \cdot \pi_c^{1/e_f} \quad (3.3)$$

$$P_{r\ exit\ ideal} = P_{r\ inlet} \cdot \pi_c \quad (3.4)$$

where π_c is the total pressure ratio across the compressor. With the reduced pressure and a fuel fraction of 0, the FAIR subroutine (described in more detail in section 3.2.1) is called to obtain all remaining fluid properties (including enthalpy (h)) at compressor exit.

A slightly different variation is employed to get the total pressure ratio across the turbine (π_{tb}). Assuming the conditions at burner exit (station 4) are known, the temperature ratio across the turbine (τ_{tb}) is identified using the required power across the shaft. The enthalpy drop across the turbine is calculated:

$$h_{tb\ exit} = h_{tb\ inlet} \tau_{tb} \quad (3.5)$$

The enthalpy at turbine exit ($h_{tb\ exit}$) is then passed into FAIR to obtain turbine exit conditions. This allows the calculation of the turbine pressure ratio:

$$\pi_{tb} = \left(\frac{P_{r\ tb\ exit}}{P_{r\ tb\ inlet}} \right)^{1/e_{tb}} \quad (3.6)$$

With the pressure ratio across the turbine we can get the ideal conditions using FAIR if we first calculate the reduced pressure:

$$P_{r\ tb\ exit\ ideal} = \pi_{tb} P_{r\ tb\ inlet} \quad (3.7)$$

Using the exit ideal and non-ideal conditions, the adiabatic efficiency is calculated for both the compressor and the turbine using total enthalpy differences:

$$\eta_c = \frac{h_{t\ ex\ ideal} - h_{t\ in}}{h_{t\ ex\ actual} - h_{t\ in}} = \frac{\dot{W}_{in\ ideal}}{\dot{W}_{in\ actual}} \quad (3.8)$$

$$\eta_{tb} = \frac{h_{t\ in} - h_{t\ ex\ actual}}{h_{t\ in} - h_{t\ ex\ ideal}} = \frac{\dot{W}_{out\ actual}}{\dot{W}_{out\ ideal}} \quad (3.9)$$

The adiabatic efficiency is stored as an output for on-design conditions, and is used in the off-design conditions.

NPSS takes a different approach to calculating the properties across compressors and turbines. As with AEDsys, inlet properties are known and the user specifies the pressure ratio across the component. The total pressure (P_t) at exit is calculated as

$$P_{t\ exit} = P_{t\ inlet} \cdot \pi_c \quad (3.10)$$

or

$$P_{t\ exit} = \frac{P_{t\ inlet}}{\pi_{tb}} \quad (3.11)$$

However, in order to obtain the enthalpy at exit, the ideal fluid conditions must first be calculated. The thermodynamic packages in NPSS requires two properties besides the fuel fraction when calculating all other fluid properties. Having the ideal total pressure, we still need one more ideal fluid attribute to obtain all fluid properties. In an ideal component there is no entropy change from inlet to exit so the fluid property solution can be obtained using the inlet entropy and calculated total exit pressure from equation 3.10 or 3.11. Non-ideal total enthalpy at component exit can then be calculated using the enthalpies and user input adiabatic efficiencies:

$$h_{t\ exit\ c} = \frac{h_{t\ ex\ ideal} - h_{t\ inlet}}{\eta_c} + h_{t\ inlet} \quad (3.12)$$

$$h_{t\ exit\ tb} = h_{t\ inlet} - (h_{t\ inlet} - h_{t\ ex\ ideal}) \cdot \eta_{tb} \quad (3.13)$$

This equation is the same as Equation 3.9; it has only been re-arranged. Subsequently, the thermodynamics routine is called using the non-ideal exit total pressure and total enthalpy as calculated above to obtain the fluid attributes.

Once the actual inlet and exit properties are known the polytropic efficiency is calculated.

$$e_c = \frac{R_{t\ inlet} \cdot \ln(\pi_c)}{R_{t\ inlet} \cdot \ln(\pi_c) + S_{exit} - S_{inlet}} \quad (3.14)$$

$$e_{tb} = \frac{R_{t\ inlet} \cdot \ln(\pi_{tb})}{R_{t\ inlet} \cdot \ln(\pi_{tb}) + S_{exit} - S_{inlet}} \quad (3.15)$$

There are two reasons to match the adiabatic efficiency between the two programs: first, the adiabatic (isentropic) efficiency is implemented by both programs identically using some form of equation 3.8. The second is that both programs utilize this efficiency for off-design calculations.

When the fluid properties at component exit are being calculated, small variations in the thermodynamic codes will result in a fluid properties discrepancy between programs. The resulting differences in the entropy across the component causes a small change in calculated efficiencies and a small temperature divergence. Small differences in temperature between the two programs can result in sizeable variations of reported power since, as Çengal and Boles put it [27]:

$$-\dot{W} = \dot{m} [(h_2 - h_1) - (V_2^2 - V_1^2) - g(z_2 - z_1)] \quad (3.16)$$

and enthalpy (h) not only dominates the other terms in the power equation, but is also a function of temperature.

To better understand how the small temperature variations between NPSS and AEDsys result in a difference of thrust, we must take a closer look at the thermodynamic models that are the basis of the two programs.

3.2.1 Thermodynamic Models. Two types of error might be encountered when comparing thermodynamic codes. As shown in Figure 3.3, the first type of error

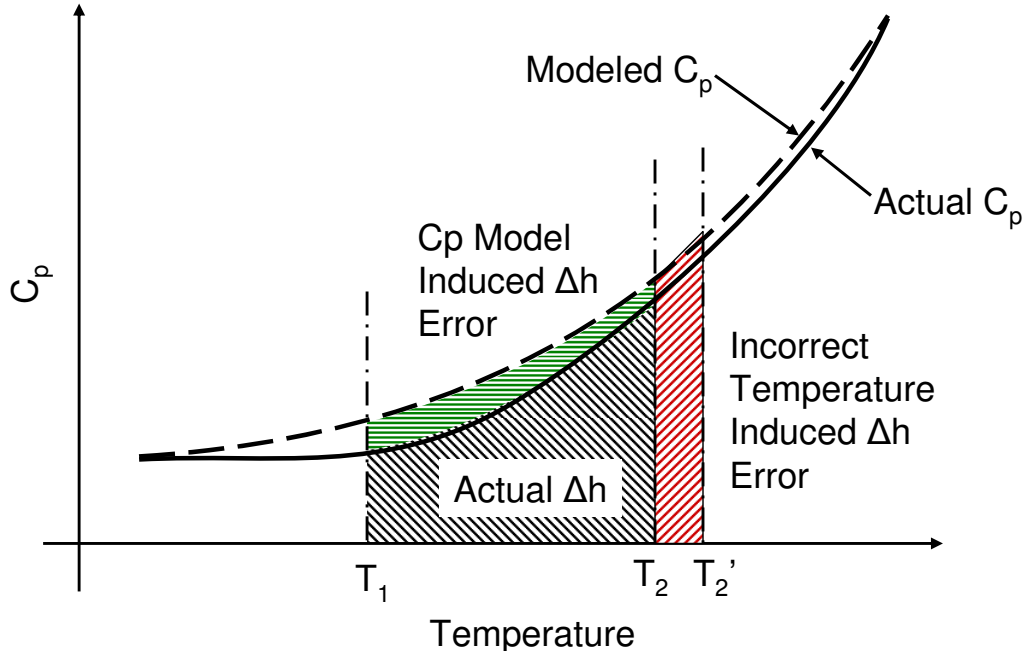


Figure 3.3: Two ways that errors might be introduced into a solution through use of the thermodynamics package.

is introduced when the specific heat model deviates from the true characteristics of the gas or other models. The area between the specific heat models is integrated to obtain the enthalpy at the given temperature using the formula:

$$h(T) = \int_T^{T_{ref}} C_p dT \quad (3.17)$$

Thus, any variation included in the integrated specific heat will yield a change in the area. Changing the area results in changing the enthalpy.

The second type of error is generated when temperature calculations across components report discrepant results. When this happens, the specific heat is integrated over different intervals. So, even if the thermodynamic model is shared the enthalpy will differ. The lesson here is that small alterations in specific heat modeling and fractions of a degree can translate into noticeable differences in final results.

To better understand the deviations in baseline engine results between AEDsys and NPSS, the thermodynamic packages of each were evaluated in order to mea-

Table 3.6: The coefficients used for the FAIR subroutine. FAIR provides the variable specific heat thermodynamic cycle calculations in AEDsys.

Pure Air		Vitiated Air	
A ₀	2.5020051x10 ⁻¹	A ₀	7.3816638x10 ⁻²
A ₁	-5.1536879x10 ⁻⁵	A ₁	1.2258630x10 ⁻³
A ₂	6.5519486x10 ⁻⁸	A ₂	-1.3771901x10 ⁻⁶
A ₃	-6.7178376x10 ⁻¹²	A ₃	9.9686793x10 ⁻¹⁰
A ₄	-1.5128259x10 ⁻¹⁴	A ₄	-4.2051104x10 ⁻¹³
A ₅	7.6215767x10 ⁻¹⁸	A ₅	1.0212913x10 ⁻¹⁶
A ₆	-1.4526770x10 ⁻²¹	A ₆	-1.3335668x10 ⁻²⁰
A ₇	1.0115540x10 ⁻²⁵	A ₇	7.2678710x10 ⁻²⁵
h _{ref}	-1.7558886 $\frac{Btu}{lbm}$	h _{ref}	30.58153 $\frac{Btu}{lbm}$
ϕ _{ref}	0.0454323 $\frac{Btu}{lbm \text{ deg } R}$	ϕ _{ref}	0.6483398 $\frac{Btu}{lbm \text{ deg } R}$

sure and correct any mismatches in reported enthalpy. The AEDsys thermodynamic package is a subroutine termed FAIR. FAIR is an 8th order polynomial fit to JANAF specific heat data for pure air, and CEA data for vitiated air according to Mattingly [28, 89-91]. First recorded by Capt. McKinney [29, 30], enthalpy and entropy functions are calculated using the relationships

$$C_p = A_0 + A_1T + A_2T^2 + A_3T^3 + A_4T^4 + A_5T^5 + A_6T^6 + A_7T^7 \quad (3.18)$$

$$h = h_{ref} + A_0T + \frac{A_1}{2}T^2 + \frac{A_2}{3}T^3 + \frac{A_3}{4}T^4 + \frac{A_4}{5}T^5 + \frac{A_5}{6}T^6 + \frac{A_6}{7}T^7 + \frac{A_7}{8}T^8 \quad (3.19)$$

$$\phi = \phi_{ref} + A_0 \ln T + A_1T + \frac{A_2}{2}T^2 + \frac{A_3}{3}T^3 + \frac{A_4}{4}T^4 + \frac{A_5}{5}T^5 + \frac{A_6}{6}T^6 + \frac{A_7}{7}T^7 \quad (3.20)$$

the same coefficients used for the calculation of specific heat (C_p) are also used for calculation of enthalpy (h) and entropy function (ϕ). The coefficients are recorded in Table 3.6. Since FAIR was not available as a stand alone routine a simplified routine titled unFAIR was written using Fortran90. The unFAIR subroutine is less robust than FAIR since it is only able to calculate the fluid properties if the temperature is

passed in. Fluid property output matches the tabulated data reported by Mattingly [28, 812-813]. A copy of this subroutine may be found in Appendix B.

NPSS is more flexible than AEDsys. It has several thermodynamic packages available [24, sec 4.1]. Four mentioned in this study are:

- GasTbl, Package developed by Pratt & Whitney based on Therm, but adding humidity calculations and some chemical equilibrium capabilities.
- allFuel, Package developed by General Electric that contains gas properties and fuel properties.
- janaf, Implementation of the National Institute of Standards and Technology gas properties prepared by Honeywell
- CEA, Implementation of NASA's Chemical Equilibrium code; however, it is not a full implementation and lacks the capability to solve detonation.

Unlike AEDsys, NPSS is designed to allow the user to control the engine architecture and implementation of controls. Responsibility for configuration does not come without a price: more control given requires more control be exercised. Careful attention to the inputs and a good understanding of the thermodynamic models should prevent the generation of incorrect solutions that appear to be valid.

In order to understand the difference in results between AEDsys and NPSS, a study was performed to measure the variations between the two programs' thermodynamic subroutines. Since specific heat is the fundamental property that defines the other thermodynamic values, the specific heat was differenced between the AEDsys and NPSS thermodynamic routines.

Figure 3.4 shows differences of the NPSS solution from the AEDsys solution for specific heat in pure air (no fuel added) for conditions at 1 atm and 40.8 atm. Above 2000 °R and 1 atm the GasTbl and allFuel values diverge from FAIR, janaf, and CEA to approximately 15% difference at 4000 °R. There is a slight difference between the two plots at the higher temperatures: note that the high-pressure data from NPSS

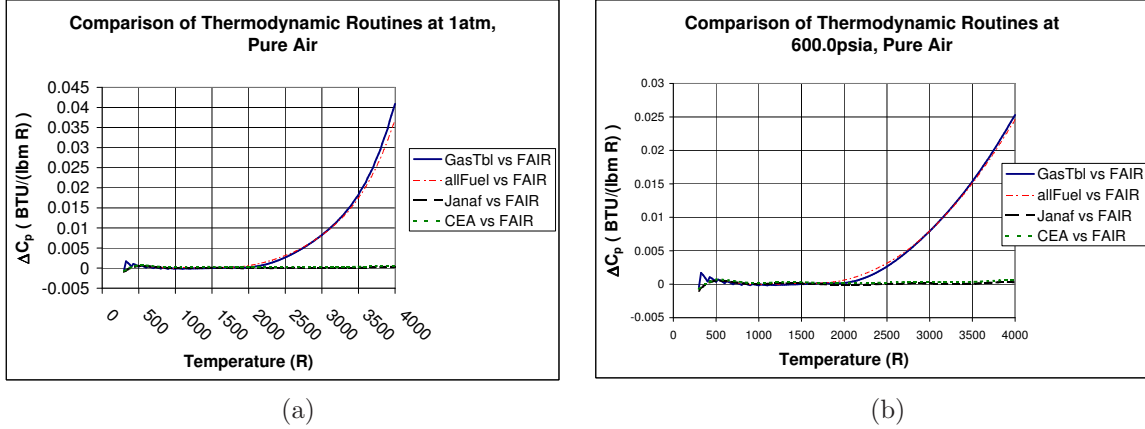


Figure 3.4: Differences between FAIR and four thermodynamic routines found in NPSS for specific heat of pure air.(a) 1 atm (b) 40.8 atm. ($C_p_{NPSS} - C_p_{FAIR}$)

is closer to the FAIR data. Below 2000 °R the variations are small, with the peak at 500 °R representing a 2% difference.

The differences in the specific heat indicate differences in enthalpy as well. Since there was no documentation about the specific relationship between enthalpy and specific heat, it was decided to plot enthalpy differences as well. Figures 3.5 and 3.6 show that for pure air, the enthalpies calculated between the two programs diverge (with the exception of janaf.) The change in enthalpy appears to be minor, but by looking at the change in Δh that occurs between 518 °R and 598 °R (temperature change across the fan and low-pressure compressor), the difference in computed power can be calculated. If the Δh values were constant (and the line shown in Figure 3.6 had no slope) then the programs would suffer from a simple bias that would have very little effect upon the power calculation (such is the case of janaf above about 1500 °R for pure air). Consider the GasTbl enthalpy change across the fan and low pressure compressor: the changing Δh value between 525 and 600 °R results in a difference between the enthalpy change ($h_2 - h_1$) calculations of 0.05 BTU/lbm. Recall that power for a compressor might be written as:

$$-\dot{W} = \dot{m} [(h_2 - h_1) - (V_2^2 - V_1^2) - g(z_2 - z_1)] \quad (3.21)$$

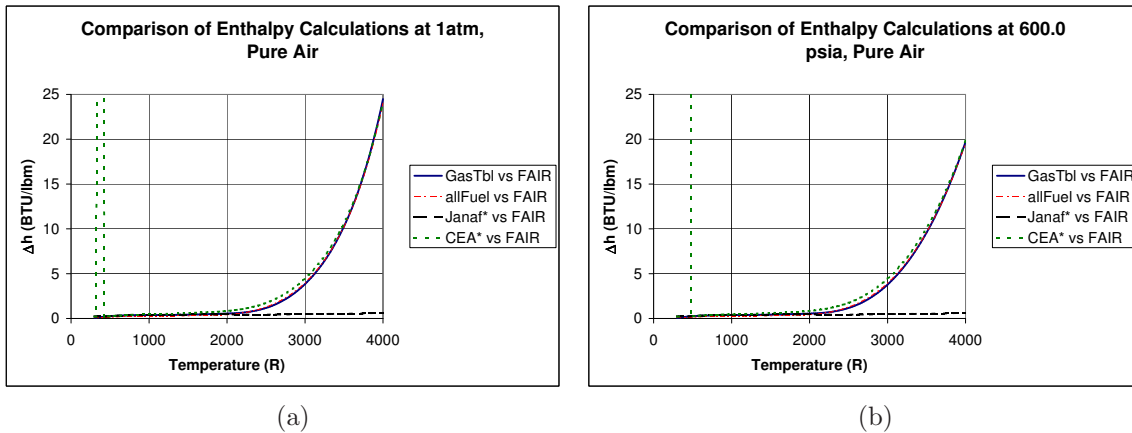


Figure 3.5: Difference between the NPSS calculation of enthalpy and FAIR calculation at (a) 1 and (b) 40.8 atm. FAIR calculates h independent of pressure, but is in better agreement when the NPSS calculations use high pressures.

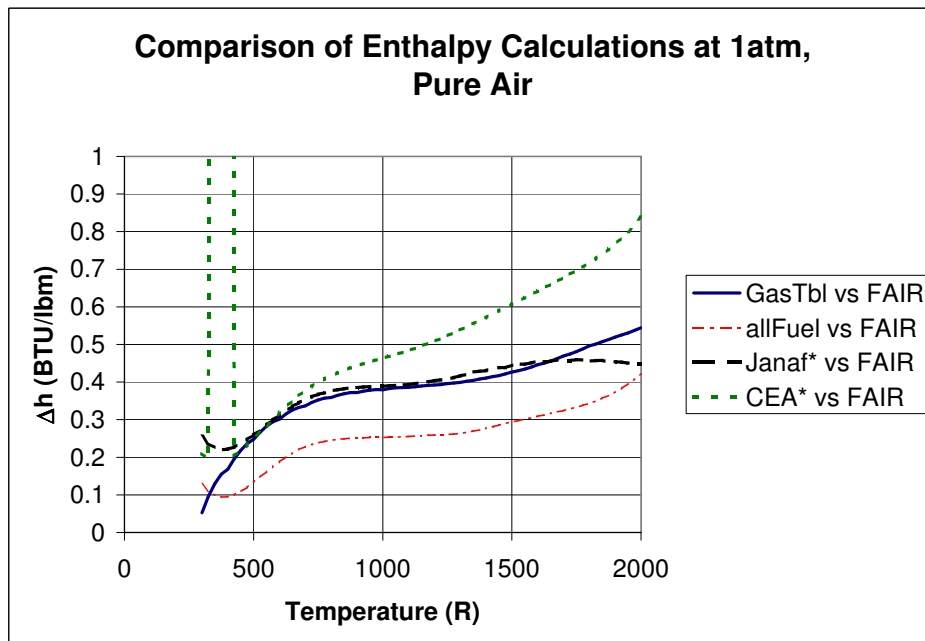


Figure 3.6: Detailed plot of the differences between FAIR and NPSS thermodynamics package calculation of enthalpy for un-reacted air at 1 atm. Although the changes are very small, when they are multiplied by a large mass flow, they can create noticeable deviations.

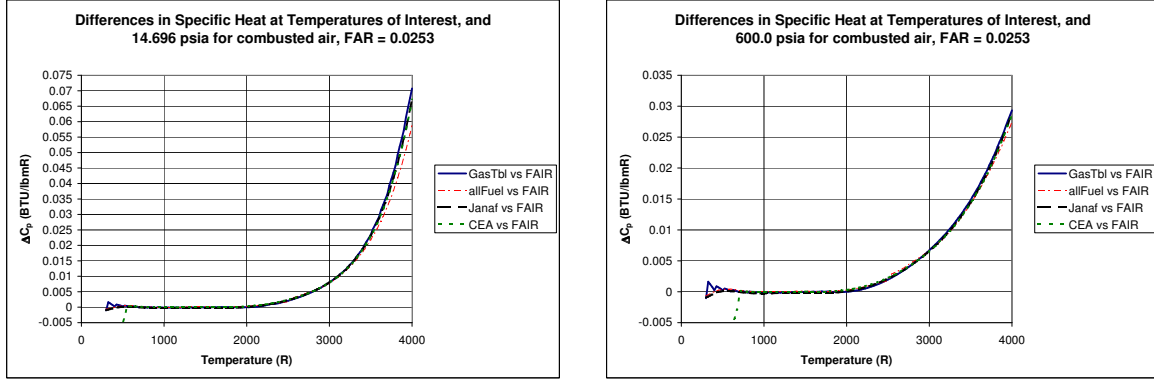


Figure 3.7: Difference in specific heats for combusted air (FAR = 0.0253) between FAIR and AEDsys routines at (a) 1 atm and (b) 40.8 atm.

so the difference in computed power becomes:

$$-\dot{W} = \dot{m} (\Delta h_2 - \Delta h_1) \quad (3.22)$$

where Δh is the difference in enthalpy between NPSS and AEDsys. For our low pressure compressor the change in power might be approximated as:

$$-\dot{W} = 1500 \frac{\text{lbm}}{\text{s}} (0.301082 - 0.266627) \frac{\text{BTU}}{\text{lbm}} = 51.68 \frac{\text{BTU}}{\text{s}} = 73.12 \text{hp} \quad (3.23)$$

For the high pressure compressor, the temperature rises from approximately 600 °R to 1425 °R, and the $-\dot{W}$ is computed as

$$-\dot{W} = 166.67 \frac{\text{lbm}}{\text{s}} (0.414196 - 0.301082) \frac{\text{BTU}}{\text{lbm}} = 18.98 \frac{\text{BTU}}{\text{s}} = 26.85 \text{hp} \quad (3.24)$$

We note that the errors in computed power will be greatest where the slope of the Δh line is the greatest, and that even relatively small differences in calculated enthalpy can be magnified by a large mass flow through the component.

Total property calculations for combusted gases follow similar trends as pure air. Data was run for a FAR of 0.0253, and the difference between specific heats is shown in Figure 3.7. Here the differences at high temperatures is notable because the

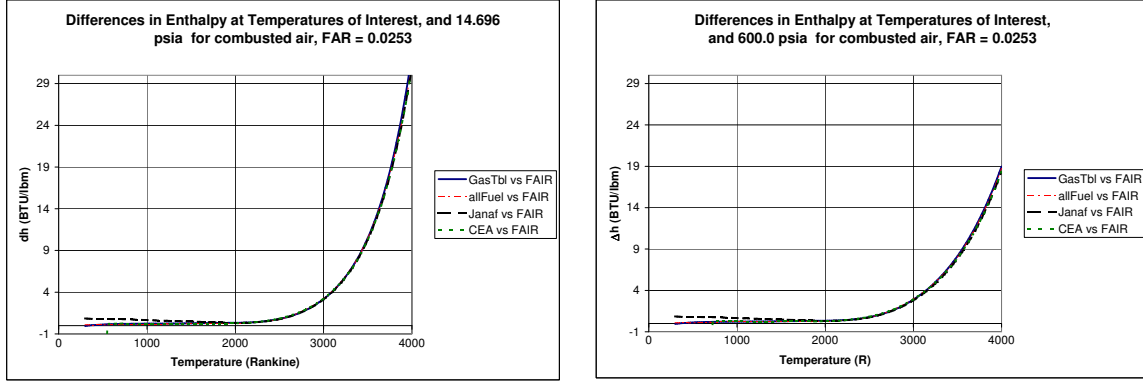


Figure 3.8: Differences in calculated enthalpies for combusted air with a FAR of 0.0253 at (a) 1 atm and (b) 40.8 atm. Agreement between the programs is much better when NPSS enthalpies are calculated using high pressure

combusted air is often above the 2000 °R point. The extreme difference at 4000 °R, 1 atm represents an 18.2% difference between the two programs. At 3000 °R, where the peak engine temperatures for this study were limited, the C_P is approximately 2.5% different. Even more concerning is that this trend is repeated in the enthalpy differences as seen in Figure 3.8. Again, when the NPSS data is generated assuming a high pressure, the data at high temperatures enjoys better agreement. At 3000 °R, there is approximately 5% reduction in the differences between the programs.

Since the baseline turbofan engines operate with a mass fuel-to-air ratio of approximately 0.0253, we can use this data to calculate differences in turbine power as well as compressor power. For the high pressure turbine work is performed between 2825 and 2125 °R. Our power difference between NPSS using GasTbl and AEDsys' FAIR is calculated as:

$$-\dot{W} = 160.42 \frac{lbm}{s} (0.358752 - 1.9993) \frac{BTU}{lbm} = -263.176 \frac{BTU}{s} = -372.3hp \quad (3.25)$$

For the low pressure turbine, work is calculated between 2075 and 1475 °R. The power difference between the programs for this temperature range is:

$$-\dot{W} = 160.42 \frac{lbm}{s} (0.264505 - 0.344959) \frac{BTU}{lbm} = -13.577 \frac{BTU}{s} = -19.21hp \quad (3.26)$$

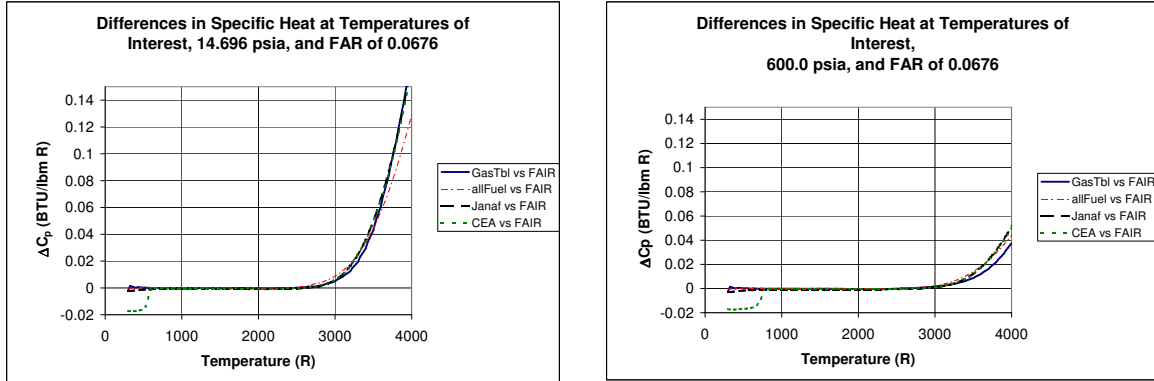


Figure 3.9: Difference in specific heats for combusted air (FAR = 0.0676) between FAIR and AEDsys routines at (a) 1 atm and (b) 40.8 atm.

By averaging the power extracted at the compressor with the power generated at the turbine, we obtain an average power difference across a given shaft due to thermodynamic model differences. From the preceding analysis, we see that AEDsys calculated approximately 50 hp net less power on the low pressure shaft and 200 hp net less power done on the high-pressure shaft. If this were the only thermodynamic difference between the two programs, it would be possible to apply the net power difference to the AEDsys shaft to achieve a net thrust that was virtually identical to that of the NPSS calculation. However, it will be shown in the results section that the AEDsys results are lower than NPSS results to begin with. This is due to the other thermodynamic difference - calculation of the temperature across a component, as was discussed in 3.2.

It is also interesting to note that for an equivalence ratio near one, the thermodynamic models again follow the same trend. As seen in figures 3.9 and 3.10, the quantities match each other more closely at low temperatures and high pressures.

3.3 Hybrid Turbofan - Pulsed Detonation Combustor Engine

3.3.1 Turbofan Configuration. The hybrid model contains the compressors, fans, and turbines of the baseline high bypass turbofan, but the burner section is replaced with a pulsed detonation combustor as seen in Figure 3.11. As much

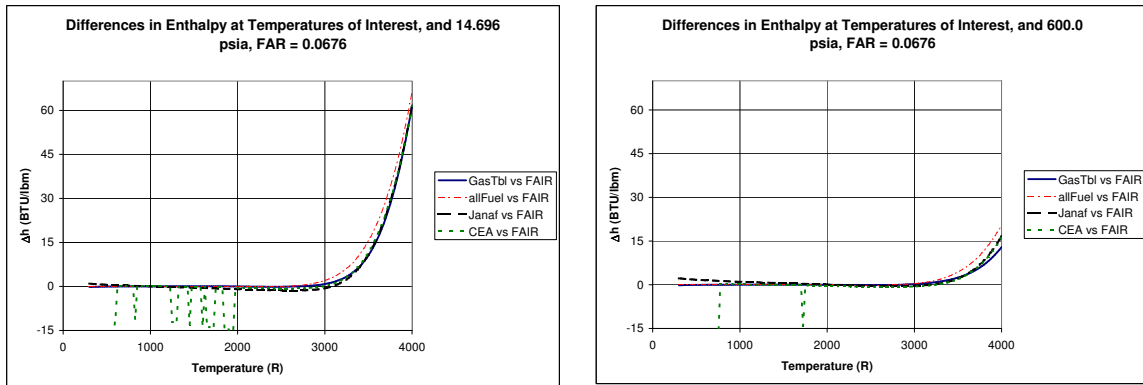


Figure 3.10: Differences in calculated enthalpies for combusted air with a FAR of 0.0676 at (a) 1 atm and (b) 40.8 atm. Agreement between specific heat models is much closer when NPSS properties are calculated using high pressure.

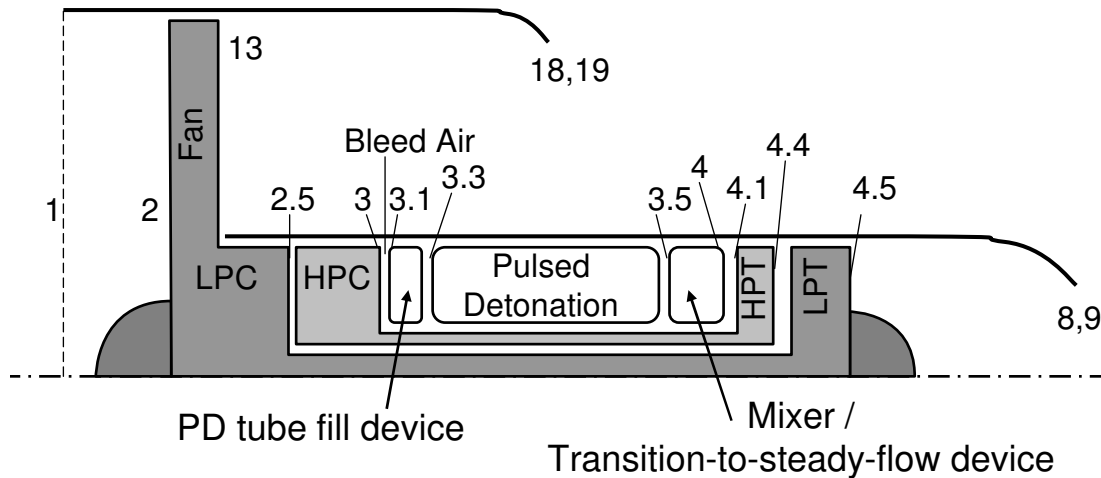


Figure 3.11: Configuration of the hybrid engine evaluated in this thesis.

of the baseline configuration as possible was retained so that the change in engine performance could be measured. The flow control at the detonation tube inlet was modeled as a pressure loss $\frac{\Delta P}{P}$ term between inlet and detonation tubes. This pressure loss matched the dry-duct pressure loss experienced by the conventional combustor (which is overly optimistic for the PDC). Mass inlet and bypass ratios were maintained constant between the hybrid and baseline models to mask the effect of thrust augmentation due to increased bypass flow. Compressor pressure ratios were also held constant, causing shaft power to remain identical to the baseline engine for all results with the exception of one of the trade off studies.

A mixer was introduced to combine the tube and internal bypass flows exiting the combustor. Utilizing the mixer introduced some difficulties since it occasionally allows NPSS to arrive at an incorrect converged solution without generating an error. The erroneous solution can be identified by a negative mach number for the secondary incoming flow (the internal bypass flow for this model). All solutions included in this thesis were checked for a correct mixer solution, with no data being reported for cycle solutions with incorrect mixer data.

3.3.2 Pulsed Detonation Combustor. The mathematics that define the detonation within the PDC model are taken from the work of Heiser and Pratt [4] with a Dyer and Kaemming correction [5]. The detonation occurs cyclically, introducing an unsteady flow. In order to assume steady flow through the same turbines as the baseline engine, a correction was made for transitioning the pulsing flow to steady state flow. No attempt was made to define an architecture for accomplishing the transition, instead, an arbitrary adjustment was made to the total temperature and pressure to account for the thermodynamic changes effected by the transition.

The pulsed detonation combustor was conceptually divided into three sections as shown in Figure 3.12:

- Flow control at detonation tube inlet

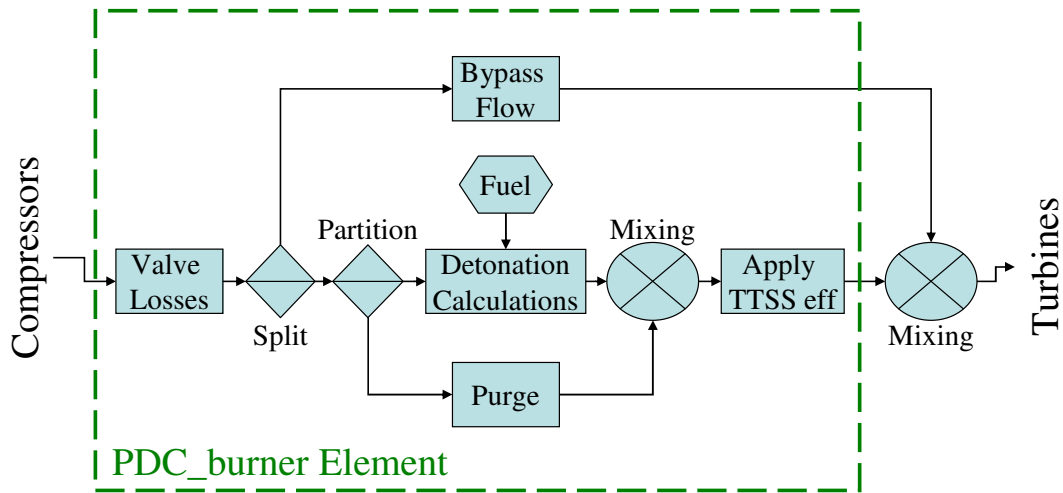


Figure 3.12: Configuration of the pulsed detonation combustor used for the hybrid engine.

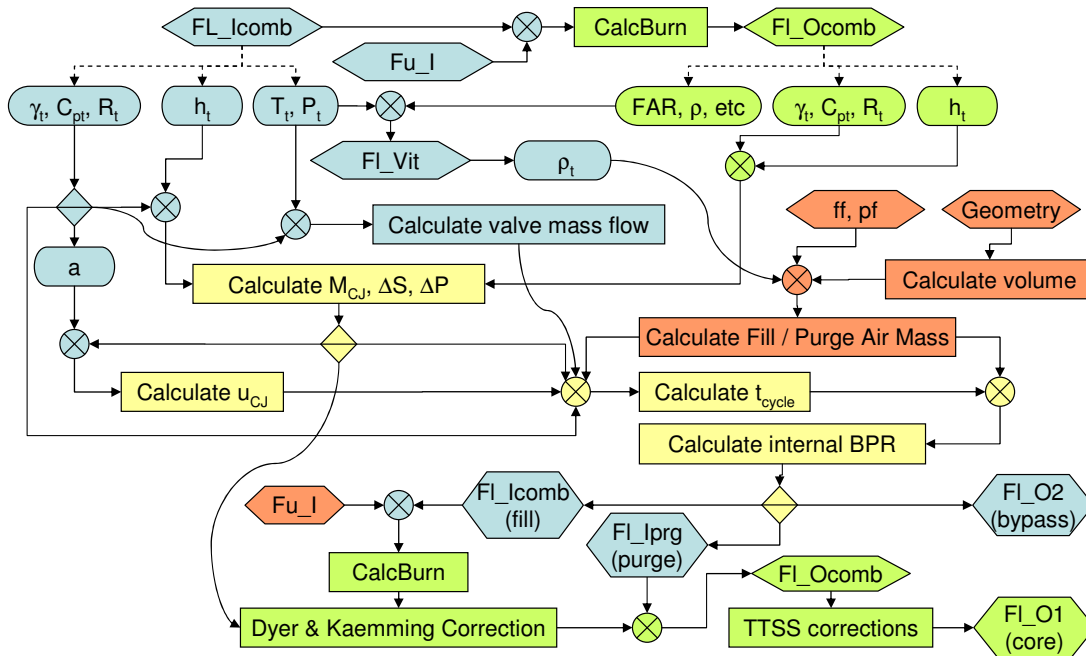


Figure 3.13: Configuration of the pulsed detonation combustor used for the hybrid engine.

- Detonation tubes
- Transition to steady state and bypass mixing device

Figure 3.12 shows how the fluid flows through the pulsed detonation combustor, but not how the information is passed through the code that calculates the fluid properties at entry and exit. The NPSS subroutine or “Element” for the PDC is included in Appendix C. Most of the code is borrowed from the NPSS “burner.int” file included in NPSS release 1.6.3. Major modifications to support the modeling pulsed detonation combustion occur after line 750.

A diagram of the information flow through the PDC_burner algorithm is shown in Figure 3.13. Although it appears complex, the diagram is meant to illustrate how the properties of the fuel an inlet flow are used to calculate the Chapman-Jouguet detonation solutions. This information is then combined with geometry inputs to determine mass flow and cycle times. Once mass flow is determined, the energy-conserved PDE solution is calculated and applied to the portion of the mass flow allocated to the detonation process. All fluids are mixed and corrected for the transition to steady state.

3.3.2.1 Detonation Properties. Central to the processes of the pulsed detonation combustor is detonation. The model used for this thesis calculates fluid properties behind the shock using the thermodynamic model described by Heiser and Pratt [4]. Implementation into NPSS required a few modifications to equations 2.7 to 2.9. This was accomplished by re-arranging the quantity $\frac{\tilde{q}}{\Psi}$ as follows:

$$\left(\frac{\tilde{q}}{\Psi}\right) = \frac{\left(\frac{q_{supplied}}{C_p T_0}\right)}{\left(\frac{T_3}{T_0}\right)} = \frac{h_4 - h_3}{C_p \cdot T_3} \quad (3.27)$$

This allowed the PDC subroutine to utilize the component inlet temperature instead of trying to find the temperature at engine inlet. The modified equation uses specific heat and enthalpies $(\gamma, C_p, h_4 - h_3)$ from both the conditions that are given

and those that are solved for. Using the solved properties in the solution requires either iteration, or pre-calculation of the combustion.

Having calculated the $\left(\frac{\tilde{q}}{\tilde{\Psi}}\right)$ value, M_{CJ} is calculated using equation 2.7.

$$M_{CJ}^2 = (\gamma + 1) \left(\frac{\tilde{q}}{\tilde{\Psi}}\right) + 1 + \sqrt{\left[(\gamma + 1) \left(\frac{\tilde{q}}{\tilde{\Psi}}\right) + 1\right]^2 - 1} \quad (3.28)$$

The Chapman-Jouguet Mach number is then used to calculate entropy gain and pressure rise across the shock using equations 2.8 and 2.9.

$$\frac{s_4 - s_3}{C_p} = -\ln \left[M_{CJ}^2 \left(\frac{\gamma + 1}{1 + \gamma M_{CJ}^2} \right)^{\frac{\gamma+1}{\gamma}} \right] \quad (3.29)$$

$$\frac{p_4}{p_0} = \frac{1 + \gamma M_{CJ}^2}{\gamma + 1} \frac{p_3}{p_0} \geq 1 \quad (3.30)$$

The forms of the equations are fairly simple, and give rise to the question of their accuracy. Glassman [7, p 248] provides M_{CJ} data plotted as a function of equivalence ratio (ϕ) at standard pressure and temperature for a variety of fuels. Figure 3.14 shows the Chapman-Jouguet Mach number calculated using the closed form solution and NPSS thermodynamic routines at standard temperature and pressure overlaid on the data from Glassman. The calculated properties utilize fuel properties similar to those of Jet-A fuel. It will be noted that the GasTbl thermodynamic package is limited to results whose equivalence ratio is less than unity, so the allFuel package was used to generate results for an equivalence ratio greater than one. The M_{CJ} was shown in these charts to be marginally above values for a large hydro-carbon such as octane (C_8H_{18}). To determine whether this discrepancy is conservative or not requires an examination of what the M_{CJ} value is used for. The examination occurs through the calculation of entropy gain. Entropy gain is one of the two fluid properties input to solve for pressure and temperature. Figure 3.15 shows how entropy error has a large effect on pressure and a negligible effect on temperatures. Also, a larger Mach will result in a larger entropy gain, pushing us to the right on this figure. Lower

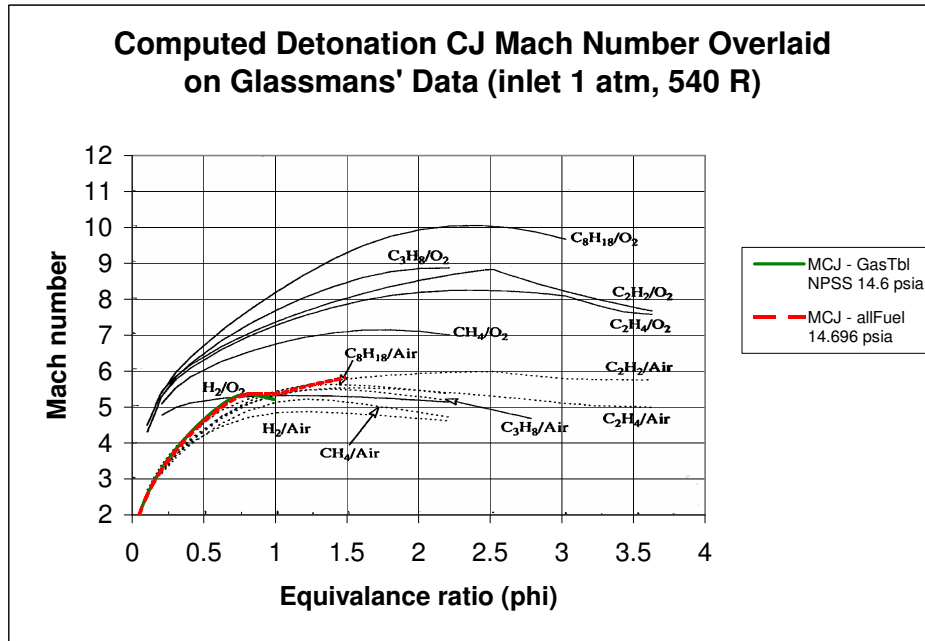


Figure 3.14: Chapman-Jouguet Mach number data overlaid on the chart from Glassman. [7, 248].

pressure decreases system performance, and the result is a conservative estimate of net thrust and TSFC.

Since the PDC is embedded in a turbofan, it experiences detonation at higher pressures than exist for the reviewed data [7, p248-250]. Figure 3.16 shows that M_{CJ} increases with increasing pressure. This is expected since increasing pressure decreases the mean free path between reacting particles, allowing for a faster transmission of energy. The calculated Mach number and the data from Glassman agree fairly well with experimental pulsed detonations, which see a detonation wave move at a Mach number slightly above 5 at standard conditions.

Having solved for pressure and entropy at detonation tube exit (station 4), it is possible to solve for all the all properties of the fluid at this point through thermodynamic relationships. However, Dyer and Kaemming [5] noted that this would be inaccurate since it ignores the eventual pressure loss that the gas will go through due to expansion waves. Figure 3.17 contains a T-S diagram explaining their proposed correction to the cycle. The expansion also results in the lowering of kinetic energy.

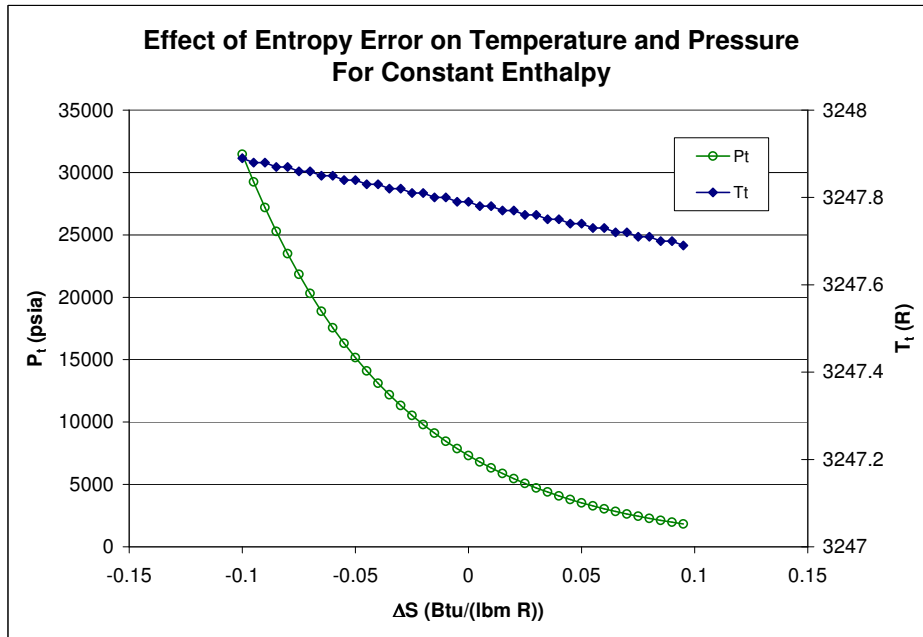


Figure 3.15: Small errors in calculation of the entropy strongly affect the pressure, but barely affect the temperature when all fluid properties are solved for using entropy and enthalpy.

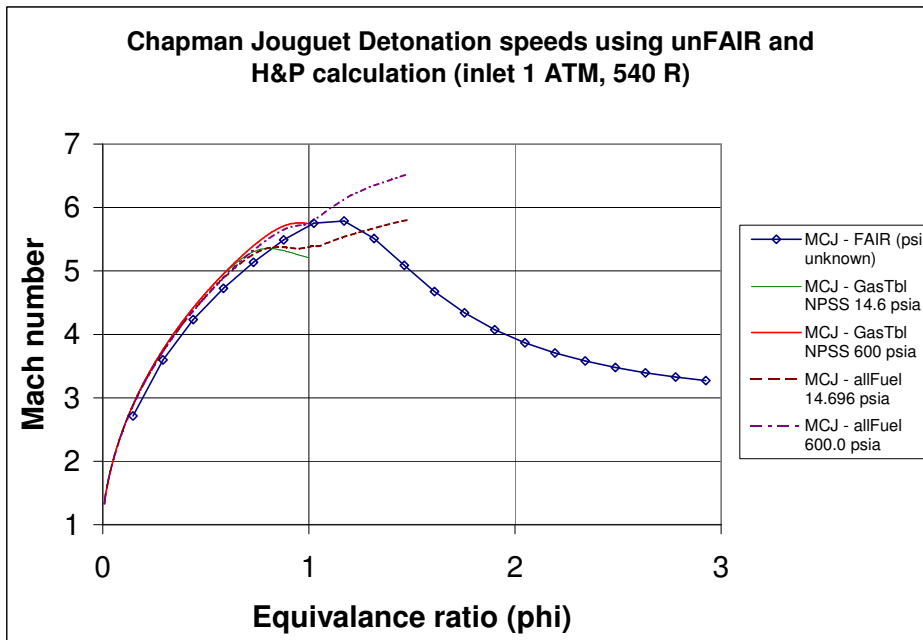


Figure 3.16: Chapman-Jouguet Mach number data calculated using several NPSS thermodynamics packages at standard and high pressures.

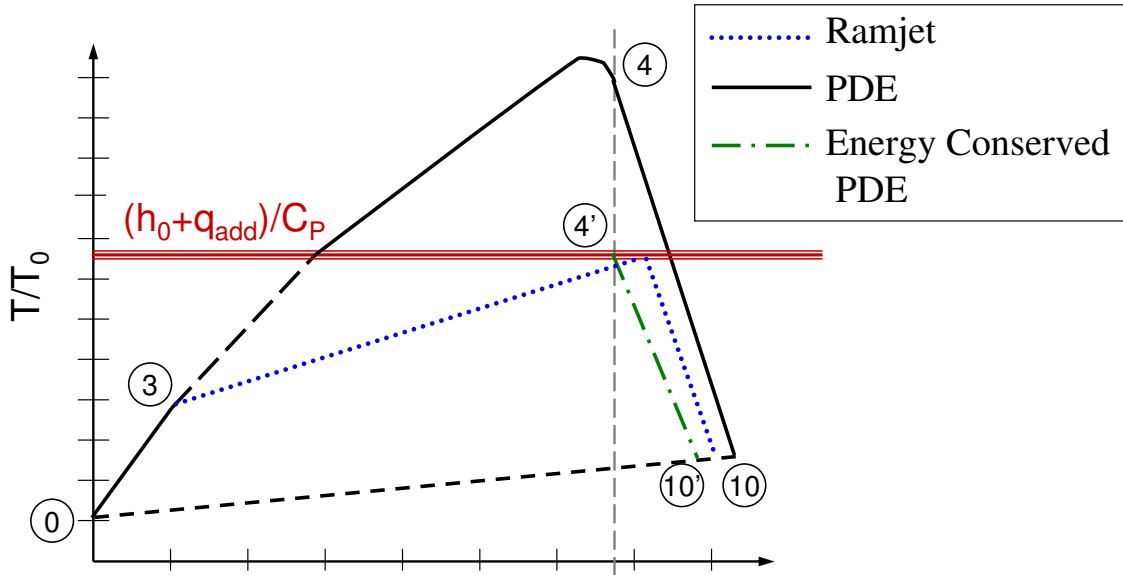


Figure 3.17: The T-S diagram shown in Dyer and Kaemming paper [5, fig 10] with the proposed correction to the PDE thermodynamic cycle. Conservation of energy will prevent incorrect calculation of entropy as the fluid exits the engine.

They proposed that the fluid properties should be solved for using the entropy gain as calculated above and the change in enthalpy liberated by the combustion process. Recall that this is the same change in enthalpy that was used to calculate $\frac{\bar{q}}{\Psi}$. This would result in a different temperature and pressure profile for the gasses at station 4. Figures 3.18 and 3.19 show the deviation that the Dyer and Kaemming [5] formulation causes to the pressure and temperature of a detonated fluid at standard conditions. The underlying data are taken from Glassman [7, pp 249-250]. It should be noted that none of the P_2/P_1 data and Heiser and Pratt [4] calculations match the pressure ratio immediately behind the shock, but overestimate the time-averaged pressure ratio over the detonation time. Dyer and Kaemming [5] values appear to under-estimate the time-averaged pressure ratio over the detonation portion of the cycle.

The pressure ratio across the shock as computed by using enthalpy and entropy should give us the correct system performance, but may not accurately represent the actual conditions in the detonation tube. Note that the pressure ratio actually becomes inverted over portions of the equivalence ratio plot for atmospheric conditions.

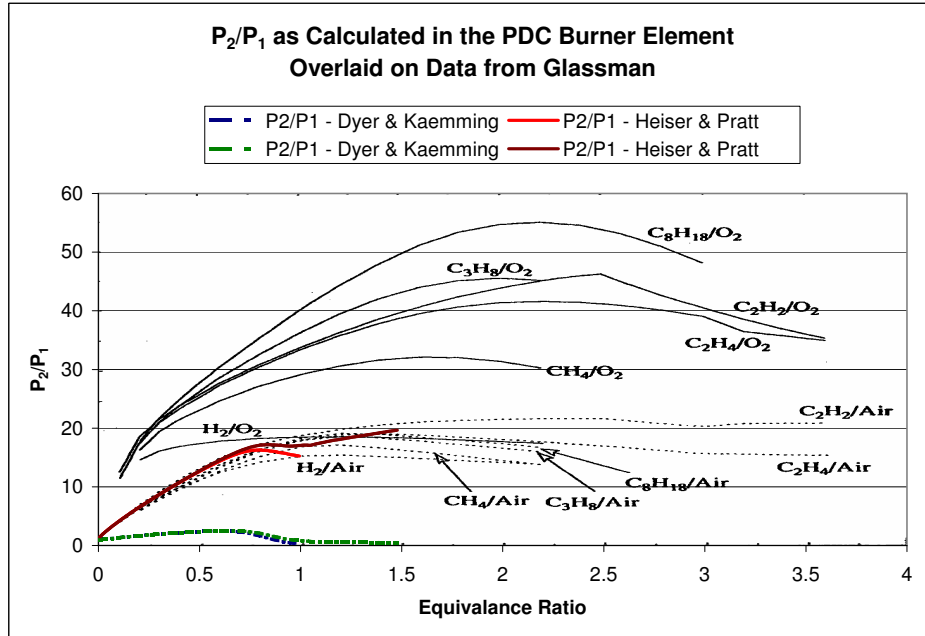


Figure 3.18: Ratio of the pressure behind to the pressure ahead of the detonation shock overlaid on data from Glassman [7, 250]. The upper line represents the pressure ratio as calculated by Heiser and Pratt, [4] the lower is the calculation using the Dyer and Kaemming [5] correction. .

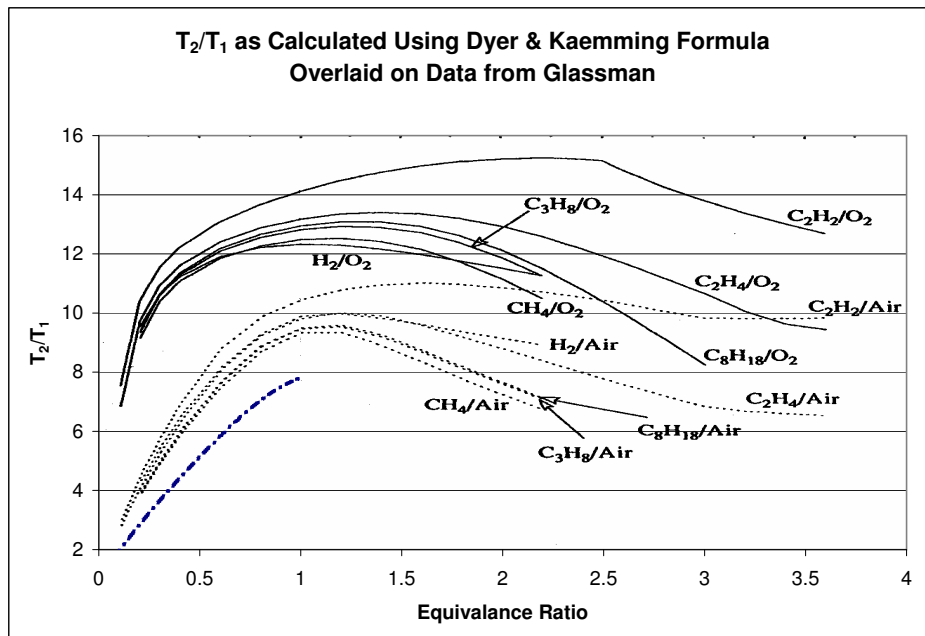


Figure 3.19: Ratio of the temperature behind to the temperature ahead of the detonation shock as calculated using the Dyer-Kaemming correction [5] . Overlaid on data from Glassman. [7, 249].

In section 3.3.2.3 we will see that we will use the pressure ratio based on the Heiser and Pratt [4] formulation to calculate the blowdown time of the system.

3.3.2.2 Determining Mass Flow Rate. Heat flow into the engine is controlled by the fuel flow and for a PDC fuel flow is defined by fill flow. PDC operation demands that the FAR be near unity for consistent detonation. Consequently the amount of air flowing into the tubes becomes the prime control of the engine. The air flowing through the detonation tubes is divided into two portions based on whether or not it is mixed with fuel. The mixed portion is termed the fill air, while the unmixed portion is called the purge air. Purge air is used for cooling of the detonation tubes while acting as a barrier between the hot combustion products of the previous and current cycle. The purge fraction pf and fill fraction ff are defined as fractions of the distance the fuel/air mixture or purge air flows relative to the tube length. Mathematically it looks like:

$$pf = \frac{l_{\text{purge air}}}{l_{\text{tube}}} \quad (3.31)$$

$$ff = \frac{l_{\text{fuel-air mixture}}}{l_{\text{tube}}} \quad (3.32)$$

where l denotes a length.

Purge and fill fractions can also be defined in terms of the volume (and subsequently mass) of the air for their respective portions of the tube filling process. Since the cross section of the fluid and the tube share the same area and the fractions are defined by the length that the respective fluids flow down the tube, the equations become:

$$pf = \frac{V_{\text{purge air}}}{A_{\text{tube}} \cdot l_{\text{tube}}} \quad (3.33)$$

$$ff = \frac{V_{\text{fuel-air mixture}}}{A_{\text{tube}} \cdot l_{\text{tube}}} \quad (3.34)$$

where A_{tube} is the cross sectional area of the tube, l_{tube} is tube length, $V_{purge\ air}$ is the purge air for one cycle, and $V_{fuel-air\ mixture}$ is the volume that the fuel-air mixture occupies in the detonation tube. Also, equation 3.34 can be re-arranged to define the volumes of purge and fill fluids in terms of tube volume and the purge/fill fractions:

$$V_{purge\ air} = pf \cdot A_{tube} \cdot l_{tube} \quad (3.35)$$

$$V_{fuel-air\ mixture} = ff \cdot A_{tube} \cdot l_{tube} \quad (3.36)$$

The purge air mass m_{purge} and fuel-air mass used for filling ($m_{fuel-air\ mixture}$) can be found using the total density (ρ_t), since the air is stopped in the tube once the valve closes:

$$m_{purge} = pf \cdot V_{tube} \cdot \rho_t \quad (3.37)$$

$$m_{fuel-air\ mixture} = ff \cdot V_{tube} \cdot \rho_t \quad (3.38)$$

These equations represent the amount of purge air and fuel-air mixture will flow into one tube during each cycle. The practice of allowing the purge and fill to sum to unity was observed when results were generated. This practice corresponds to completely filling a detonation tube each cycle without allowing any spillage or retention of hot gasses from the previous cycle.

Knowing how much air to send through the valve at the opening of the detonation tube requires separating the fuel from the air in the fill portion of the pulse-detonation cycle. This can easily be done at this point because mixed fuel air mass can be written as:

$$m_{fuel-air\ mixture} = m_{fill\ air} + m_{fuel} \quad (3.39)$$

$$\frac{m_{fuel-air\ mixture}}{m_{fill\ air}} = \frac{m_{fill\ air}}{m_{fill\ air}} + \frac{m_{fuel}}{m_{fill\ air}} \quad (3.40)$$

$$\frac{m_{fuel-air\ mixture}}{m_{fill\ air}} = 1 + FAR \quad (3.41)$$

$$m_{fill\ air} = \frac{m_{fuel-air\ mixture}}{1 + FAR} \quad (3.42)$$

$$m_{fill\ air} = \frac{ff \cdot V_{tube} \cdot \rho_t}{1 + FAR} \quad (3.43)$$

where $m_{fill\ air}$ is the mass of the air that will be detonated and m_{fuel} is the fuel used during one cycle. Once the mass of the air flowing into the tubes during one cycle is known, we can get the time averaged steady state mass flow rate ($\dot{m}_{air\ total}$) by multiplying it with the cycle frequency (f).

$$\dot{m}_{air\ total} = (m_{fill\ air} + m_{purge}) \cdot f \quad (3.44)$$

However, since the valve is only open for a fraction of the detonation cycle, the actual mass flow rate through the valve may be much greater than the steady-state depending on synchronization of multiple tubes. The maximum mass flow rate through the inlet valves must be calculated to ensure that the mass flow does not exceed the capabilities of the system. Treating the valve as a nozzle, and assuming the geometry allows for choking, we follow Mattingly [3, 9-10] and Wilson and Korakianitis [26, 70-71] in writing the mass flow rate during the valve-open time (\dot{m}_{vo}) as:

$$\dot{m}_{vo} = \frac{A_{valve} \cdot P_t}{\sqrt{T_t}} \sqrt{\frac{\gamma}{R}} MFP \quad (3.45)$$

Where MFP is the mass flow parameter that can be written as a function of Mach number, or static-to-stagnation pressure ratio:

$$MFP = M \left(1 + \frac{\gamma - 1}{2} M^2 \right)^{\frac{-(\gamma+1)}{2(\gamma-1)}} \quad (3.46)$$

$$MFP = \sqrt{\left(\frac{2}{\gamma - 1} \right) \left[\left(\frac{P_t}{P} \right)^{\frac{\gamma-1}{\gamma}} - 1 \right] \left(\frac{P_t}{P} \right)^{\frac{-(\gamma+1)}{\gamma}}} \quad (3.47)$$

This allows us to define the maximum mass flow rate for a given Mach number and valve inlet area. The flow rate during the purge and fill portions of the cycle will help

to define cycle time. More importantly, it will determine the valve-open time fraction (τ_{vo}). This parameter is important because it prescribes how many tubes need to be open at one time in order to maintain a constant flow into the PDC.

3.3.2.3 Cycle Time. Cycle time could have easily been an input for the program, but the work of Schauer et al. [21] showed that when a detonation tube is fired into a turbine the detonation cycle is changed. In order to get a more realistic feeling for what a minimum cycle time would be calculations were made based on some simple assumptions: choked flow at tube exit for blowdown, a detonation moving at M_{CJ} , and a valve flow at or below choked conditions. As described in Section 3.3.2.2, mass flow rate through the engine is calculated for the on-design condition as the mass of air to fill and purge multiplied by the frequency of filling. Frequency of filling is determined by the inverse of the cycle time (t_{cycle}):

$$f = \frac{1}{t_{cycle}} \quad (3.48)$$

Cycle time is the sum of several smaller periods of time: detonation time, blowdown time, purge time, and fill time. Detonation time is approximated as the time to initiate detonation and move the wave down the detonation tube at $M_2 = 1$, where the subscript 2 indicates the detonated fluid. Implementation requires the calculation of the velocity:

$$t_{detonation} = DDT + \frac{l_{tube}}{u_{CJ}} \quad (3.49)$$

Where DDT is the detonation-to-deflagration time, u_{CJ} is the Chapman-Jouguet detonation wave velocity and l_{tube} is the length of the detonation tube. Detonation-to-deflagration time is poorly understood, and is requested as a user input. Ideally, u_{CJ} would be calculated using the speed of sound in the combusted air. In the algorithm coded for this thesis, the properties of the detonated fluid were never completely solved for, thus, the detonation wave velocity was calculated using M_{CJ} multiplied by the speed of sound in the pre-detonation fuel-air mixture.

$$u_{CJ} = M_{CJ} \cdot \sqrt{\gamma_1 R_1 T_{t1}} \quad (3.50)$$

where the subscript 1 indicates that the fluid properties of the fuel-air mixture are taken before detonation. Again, since the pre-detonation fuel-air mixture is essentially at rest, the stagnation properties were used to calculate the speed of sound. It can easily be seen from Equation 3.49 that detonation time is directly proportional to the length of the tube.

Blowdown time ($t_{Blowdown}$) begins after the detonation shock has exited the tube. The shock leaves behind a hot compressed fluid expanding out of the end of the tube. If we assume the tube exit to be choked, then we will see an exponential decay of pressure and mass within the tube. As the mass flows through the choked throat at the end of the tube, the static properties of the gas change, ρ decreases, pressure decreases, and volume remains constant. Our basic steady-flow thermodynamic equation $\dot{m} = \rho Au$ will not allow us to calculate the amount of time it takes to discharge the gasses; this is not steady flow. Instead, we use the relationship identified in an experiment designed by Dutton and Coverdill [33] based off of the pressure ratio across the throat:

$$\Delta t = \frac{\ln\left(\frac{P_2}{P_1}\right)}{\left(\frac{\gamma+1}{2}\right)^{\frac{\gamma}{2(\gamma-1)}}} \cdot \frac{Vol}{A_t a_1} \quad (3.51)$$

As seen in Figure 3.20, blowdown time is strongly influenced by the pressure ratio. What is not shown in this plot is how little γ influences blowdown time; its effect is inconsequential over the range of possible specific heat ratios found in an air breathing engine. Significantly, when the tube exit has a constant cross section, and the exit area is the same as that cross section, the trailing term will reduce to l_{tube}/a_1 forcing the blowdown time to be directly proportional to the tube length.

Blowdown time is the most complex piece of the cycle time to calculate. It utilizes a fraction of the pressure ratio calculated using the Heiser and Pratt formulation [4, eq 18]. Figure 3.18 indicates the Heiser and Pratt formula calculates the

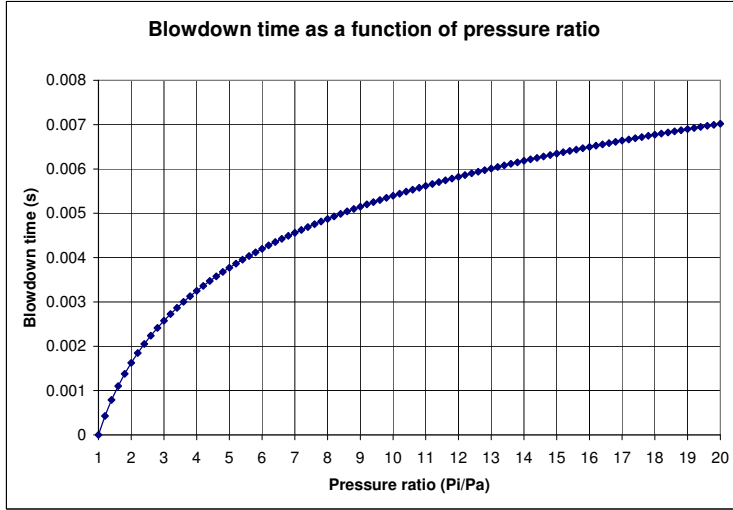


Figure 3.20: Calculated blowdown time plotted as a function of pressure ratio for a 36 inch tube and sonic velocity of 2200 ft/s.

pressure ratio corresponding to the characteristic pressure spike seen in PDE, while the Dyer and Kaemming [5] formulations generates a lower pressure ratio than is seen in the laboratory during blowdown. Attempting to err on the conservative side, we pick the over-estimate of the pressure ratio that yields a longer time. In order to keep the blowdown time reasonable, the pressure ratio is calculated as $\frac{4}{10}$ of the Chapman-Jouguet pressure rise calculated from the formulae from Heiser and Pratt [4].

Purge and fill times are closely related since they both deal with the flow of air into the detonation tube through a valve. As mentioned in subsection 3.3.2.2, we assume that the flow into the tube is constrained by flow through a valve whose throat has an area smaller than the tube cross section area. Once the mass flow rate for choked flow has been established (the maximum mass flow rate) based on a design valve inlet Mach number, purge and fill times (t_{purge} and t_{fill}) become a simple calculation:

$$t_{fill} = \frac{m_{fill\ air}}{\dot{m}_{vo}} \quad (3.52)$$

$$t_{purge} = \frac{m_{purge}}{\dot{m}_{vo}} \quad (3.53)$$

3.3.2.4 *Internal Bypass Ratio.* The configuration of the hybrid engine sets the main bypass ratio to a fixed value for all cases. This means that there is a constant mass flow rate into the core of the engine. Not all of this core flow will pass through the tubes however. Recall that only a small amount of purge air is needed to act as a buffer between the hot combusted gases of the previous cycle, and the volatile fuel-air mixture in the current cycle. If the baseline engine only required a fuel mass flow rate of 1 lbm/sec, then we expect the hybrid engine to require a similar flow rate. Since the detonation engine operates at a higher equivalence ratio than normal engines, it should require less air to mix with the fuel for a similar enthalpy generation. Balancing the mass flowing through the tubes with the fuel necessitates shunting some of the air around the detonation tubes. Ideally, this internal bypass (iBPR) air provides cooling of the tube walls, and will be mixed with the combustion products to lower the gas temperature before entering the turbines. This internal bypass might be utilized in the transition-to-steady-state device. This may not be desirable since such a connection provides a link to the compressor exit.

Mass flow for the iBPR is assumed to be steady state and, at design conditions, is defined by the formula:

$$iBPR = \frac{\dot{m}_{iBPR}}{\dot{m}_{tube\ air}} \quad (3.54)$$

where air mass flow rate through the internal bypass (\dot{m}_{iBPR}) is defined as:

$$\dot{m}_{iBPR} = \dot{m}_{core\ air} - \dot{m}_{tube\ air} \quad (3.55)$$

and where $\dot{m}_{core\ air}$ is the combined mass flow rate coming into the PDC before it is split.

No pressure loss term is applied to the iBPR because it is separated from the core flow after a dry-duct pressure loss term was applied. For the flow going into

the tubes, this dry duct pressure loss is intended to represent pressure loss through a valve.

Heat transfer from the detonation tubes to the iBPR could be accomplished through implementation of a simple heat transfer wall element in NPSS. This element is evoked after the detonation calculations are performed. Another option is to include a heat transfer element imbedded within the PDC element. Either choice will require several inputs:

- areas available to the two fluids
- film transfer coefficients
- fluid inlet properties
- wall mass
- wall specific heat

Attempts were made to include the wall heat transfer element in the model for this thesis, but NPSS solver difficulties caused this option to be pushed aside until more time can be devoted to improving the initial conditions for the solver.

3.3.2.5 Iterations. If the Chapman-Jouguet Mach number is calculated before we calculate mass flow and cycle time, we should be able to carefully craft a PDC routine that does not require iteration. Otherwise, iteration is required to balance the mass flow through the tubes for a given geometry since mass flow depends on cycle time, and cycle time depends on mass flow rate. Recall that total mass flow rate through the tubes is calculated using the air mass required for filling the tube to the designated purge and fill fractions, multiplied by the inverse of the cycle time (see equation 3.44). Fill and purge portions of the detonation cycle are calculated using the mass required to fill the tubes. Blowdown time is only a function of pressure ratio calculated using the Chapman-Jouguet Mach number, and is therefore independent of mass flow if the Mach number can be calculated at the outset. Detonation time is calculated using the DDT and the time it takes the detonation wave to traverse the

length of the tube, leaving it independent of mass flow but dependent on a detonation wave Mach number. As seen in Appendix C, the method of obtaining the shock properties for this thesis allows for a non-iterative solution.

3.3.2.6 Transition to Steady State Flow. Rasheed et al. [21] showed that exhausting a pulsed detonation combustor directly into a turbine lowers the turbine efficiency, and Rasheed et al. [18] discussed the structural ramifications affecting engine life. Since the pulsed flow into the turbine is unfavorable, the hybrid model is based on the assumption that the flow into the turbine sections is steady flow. No attempt was made to define the specific architecture that would transition the pulsed flow back to steady flow. Instead, a subelement to the PDC was created that allows for the application of a pressure drop and an enthalpy loss. Pressure loss was chosen because it translates to a diffusion process, whereas the enthalpy loss was chosen because it translates to work rates through equation 3.16.

3.3.2.7 Improvements. It has been said for centuries that “Rome was not built in a day,” and the idea it communicates holds true for any engineering endeavor. Just as improvements were made to Rome with each passing year, improvements could be made with this. Some of the improvements that should be considered are:

- Replacing the detonation wave property calculations
- Implementation of a pulsed-detonation performance map built from CFD or experimental data similar to the method proposed by Paxson, [12] and implemented by Tangirala et al. [17]
- Improving blowdown time calculations
- Calculating heat transfer to bypass air
- Calculating heat transfer to purge air

- Calculating compression of purge air due to the shock initially generated by detonation
- Adaptation of the PDC element to allow for off-design performance

It had been hoped that many of these improvements could be included in the model used for the results section, but there came a point where work on the model had to stop so that deadlines for documentation could be met. Any of these changes have the potential to shift all results by several percent, which would then change any conclusions made in this thesis.

IV. Results

This chapter contains not only the data generated using the hybrid engine modeled in NPSS, it also contains data that demonstrate the close agreement of the baseline engine solutions generated using AEDsys and NPSS. The baseline engine comparison is significant because it provides an acceptable foundation for comparing the hybrid engine.

4.1 *Comparison of Baseline Turbofan*

Both the NPSS and AEDsys models of the baseline high-bypass turbofan were evaluated at on- and off-design conditions. For on-design conditions, reports for the component analysis were generated and engine parameters were carefully matched between the two programs. The close correlation in the results indicate a reasonable understanding of the NPSS program was achieved.

4.1.1 Baseline Engine On-Design Results. Tables 4.1 and 4.2 show the results and input parameters for baseline engine on-design. The off-design point included in these tables is not of any significance, but foreshadows the divergence of the off-design solutions. On-design thrust varies by approximately 1%, and thrust-specific fuel consumption (TSFC) varies by approximately 0.8%. The NPSS results using the janaf thermodynamic package (shown in tables 4.2 and 4.4) are marginally lower than results generated in NPSS by other thermodynamic packages such as GasTbl. It will be noted that the conditions at the inlet are nearly identical and slowly diverge at each station moving through the engine. This trend is even more apparent in the component interface reports in tables 4.3 and 4.4.

4.1.2 Baseline Turbofan Comparison Off-Design. Figure 4.1 shows throttle hooks generated by NPSS and AEDsys across several altitudes and range of thrust levels. The NPSS data was generated using the GasTbl thermodynamics package since it is the easiest to implement. An attempt was made to produce a similar chart using the janaf thermodynamics package. As seen in Figure 4.2, there was some

Table 4.1: AEDsys engine test data showing the design point. It is from this report that the calculated adiabatic efficiency may be taken.

AEDsys (Ver. 4.010) Turbofan -
 Separate Exhaust, Dual Spool Date:11/22/2006 9:40:50 PM
 Engine File: C:\AFIT\MENG733\AEDsys\HBtbfan\eng13 vsh.REF

Input Constants
 Pidmax= 0.9950 Pi b = 0.9600 Eta b = 0.9950 Pi n = 0.9850
 Eta cL= 0.8827 Eta cH= 0.8573 Eta tH= 0.9057 Eta tL= 0.9084
 Eta mL= 0.9900 Eta mH= 0.9900 Eta PL= 0.9900 Eta PH= 0.9900
 Eta f = 0.8827 PTO L = 0.0KW PTO H = 105.7KW hPR = 18400
 Bleed = 1.00% Cool 1= 5.00% Cool 2= 5.00% Pi nf = 0.9800
 Control Limits: Tt4 = 3200.0 Pi c = 32.00 Tt3 = 1859.7
 Pt3 = 650.0 %N LP = 110.00 %N HP = 110.00
 ** Thrust Scale Factor = 1.0000

Parameter	Reference**	Test**
Mach Number @ 0	0.0100	0.0100
Temperature @ 0	518.67	518.67
Pressure @ 0	14.6960	14.6960
Altitude @ 0	0	0
Total Temp @ 4	2900.00	3118.30
Pi r / Tau r	1.0001/ 1.0000	1.0001/ 1.0000
Pi d	0.9950	0.9950
Pi f / Tau f	1.5600/ 1.1540	1.6943/ 1.1848
Pi cL / Tau cL	1.5600/ 1.1540	1.6943/ 1.1848
Pi cH / Tau cH	16.6667/ 2.4379	18.4128/ 2.5106
Tau m1	0.9711	0.9705
Pi tH / Tau tH	0.2324/ 0.7163	0.2329/ 0.7199
Tau m2	0.9821	0.9811
Pi tL / Tau tL	0.1991/ 0.6808	0.1737/ 0.6624
Control Limit		PIC Max
LP Spool RPM (% of Reference Pt)	100.00	109.57
HP Spool RPM (% of Reference Pt)	100.00	103.86
Alpha	8.000	7.693
Pt19/P19	1.5213	1.6522
P0/P19	1.0000	1.0000
Mach Number @ 19	0.8011	0.8819
Pt9/P9	1.1323	1.1873
P0/P9	1.0000	1.0000
Mach Number @ 9	0.4351	0.5138
Mass Flow Rate @ 0	1500.00	1667.02
Corr Mass Flow @ 0	1499.88	1666.89
Flow Area @ 0	1761.143	1957.117
Flow Area* @ 0	30.428	33.814
Flow Area @ 8 + 18	27.442	27.442
MB - Fuel/Air Ratio (f)	0.02523	0.02825
Overall Fuel/Air Ratio (fo)	0.00249	0.00289
Specific Thrust (F/m0)	27.43	30.55
Thrust Spec Fuel Consumption (S)	0.3274	0.3407
Thrust (F)	41145	50934
Fuel Flow Rate	13472	17355
Propulsive Efficiency (%)	2.47	2.23
Thermal Efficiency (%)	34.76	37.04
Overall Efficiency (%)	0.86	0.83

Table 4.2: On design engine parameters for NPSS using janaf thermodynamics in a format similar to the AEDsys engine test format. Polytropic efficiencies displayed here (e_f , e_{cL} , e_{cH} , e_{tH} , e_{tL}) are different than those input into the AEDsys program.

```

NCP          NPSS_1.6.3 - Rev: B      model:CmpareAEDsys.mdl with mixers
run by:      Ionio      solutionMode= STEADY_STATE
converge=    1 case:    1      time:    0.000  timeStep:0.0500
therm_package: Janaf iter/pas/Jac/Broy= 11/ 20/ 1/ 9 run: 11/22/06 21:51:14

```

Design Values

```

Pidmax= 0.99500  Pi b = 0.96000  Eta b = 0.99500  Pi n = 0.98500
Eta f = 0.88270  Eta cL= 0.88270  Eta cH= 0.85730  Eta tH= 0.90570  Eta tL= 0.90840
e f = 0.88982  e cL = 0.88982  e cH= 0.89984  e tH= 0.88414  e tL= 0.88912
gam2 = 1.40023  gam25= 1.39843  gam3= 1.35088  gam4= 1.27704  gam45= 1.30092
Eta mL= 0.99000  Eta mH= 0.9900  gam5=1.32930
PTO L = 0.0kW  PTO H = 106.8kW  hPr = 18400
Bleed = 1.00%  Cool 1= 5.00%  Cool 2= 5.00%  Pi nf = 0.98000
** Thrust scale DOES NOT APPLY...

```

	CASE 0: DESIGN	CASE 1: OFFDESIGN
Converged? (0 - no, 1 - yes)	1	1
Mach Number at 0	0.0100	0.0100
Temperature at 0	518.6700	518.6700
Pressure at 0	14.6960	14.6960
Altitude at 0	0.0000	0.0000
Total Temp at 4	2900.0	3118.3
Pi r / Tau r	1.0001/ 1.0000	1.0001/ 1.0000
Pi d	0.9950	0.9950
Pi f/ Tau f	1.5600/ 1.1534	1.6221/ 1.1867
Pi cL/ Tau cL	1.5600/ 1.1534	1.7073/ 1.1870
Pi cH/ Tau cH	16.6667/ 2.3727	17.4925/ 2.4219
Tau m1	0.9760	0.9757
Pi tH/ Tau tH	0.2372/ 0.7466	0.2411/ 0.7534
Tau m2	0.9849	0.9841
Pi tL/ Tau tL	0.2030/ 0.7078	0.1821/ 0.6952
LP Spool RPM (\% reference pt)	100.000	106.219
HP Spool RPM (\% reference pt)	100.000	98.957
Alpha	8.000	7.528
Pt19/P19	1.5213	1.5818
P0/P19	1.0000	1.0000
Mach Number at 19	0.7979	0.8367
Pt9/P9	1.1778	1.2340
P0/P9	1.0000	1.0000
Mach Number at 9	0.4995	0.5690
Mass Flow Rate at 0	1500.0000	1569.9409
Corr Mass Flow at 0	1499.9050	1569.8415
Flow Area at 0 (ft2)	1756.6819	1838.5912
Flow Area* at 0 (ft2)	30.3544	31.7698
Flow Area at 8 and 18 (ft2)	26.4735	26.4735
MB - Fuel/Air Ratio (f)	0.02527	0.02845
Overall Fuel/Air Ratio (fo)	0.00250	0.00297
Specific Thrust (F/m0)	28.0445	30.0347
Thrust Spec Fuel Consumption (S)	0.3248	0.3601
Thrust (Fn)	41546.2151	46607.8897
Fuel Flow Rate (lbm/hr)	13496.0711	16783.0420
Propulsive Efficiency (%)	2.4480	2.2861
Thermal Efficiency (%)	35.3755	34.1603
Overall Efficiency (%)	0.8660	0.7809

Table 4.3: Component interface for AEDsys model on design conditions.

Design Interface Quantities (Version 4.010) Date:11/22/2006 9:29:25 PM											
Filename: C:\AFIT\MENG733\AEDsys\HBtbfan\Test1h.AED											
Engine File Name: C:\AFIT\MENG733\AEDsys\HBtbfan\eng13 vsh.REF											
Thrust Scale Factor = 1.0000											
Station	\dot{m} (lbm/s)	γ	Pt (psia)	Tt (R)	P (psia)	T (R)	Mach	Velocity (ft/s)	Area (ft ²)	Area* (ft ²)	I (lbf)
0	1500.00	1.3986	14.697	518.68	14.696	518.67	0.0100	11.18	1761.14	30.428	3727487.9
1	1500.00	1.3986	14.697	518.68	9.644	460.00	0.8000	842.40	31.591	30.428	83144.5
2	1500.00	1.3986	14.624	518.68	12.330	494.06	0.5000	545.64	40.975	30.581	98186.8
13	1500.00	1.3981	22.813	598.56	19.708	573.95	0.4640	545.64	29.780	20.998	109950.3
core	166.67	1.3981	22.813	598.56	19.708	573.95	0.4640	545.64	3.309	2.333	12216.7
bypass	1333.33	1.3981	22.813	598.56	19.708	573.95	0.4640	545.64	26.471	18.665	97733.6
2.5	166.67	1.3981	22.813	598.56	19.708	573.95	0.4640	545.64	3.309	2.333	12216.7
3.0	166.67	1.3546	380.213	1425.49	357.841	1402.91	0.3015	545.64	0.445	0.218	25779.3
3.1	148.33	1.3546	380.213	1425.49	357.841	1402.91	0.3015	545.64	0.396	0.194	22943.6
3.2	148.33	1.3535	376.410	1425.49	378.031	1423.88	0.0800	145.80	1.438	0.196	78942.1
MB fuel	3.7421										
4	152.08	1.2952	365.004	2900.00	199.221	2529.99	1.0000	2376.71	0.302	0.302	19905.9
4.1	160.41	1.2952		2829.75							
4.4	160.41	1.2952	84.841	2099.67							
4.5	168.74	1.3201	84.841	2068.49	46.012	1786.21	1.0000	2016.10	1.209	1.209	18582.3
5	168.74	1.3410	16.894	1456.12	13.377	1372.29	0.6000	1068.65	6.026	5.047	17213.0
8	168.74	1.3388	16.640	1456.12	14.696	1410.99	0.4351	785.08	7.677	5.124	20364.4
18	1333.33	1.3535	22.357	598.56	14.696	530.66	0.8011	906.08	19.765	19.046	78161.6

Table 4.4: NPSS component interface data utilizing the janaf thermodynamics package.

NCP NPSS.1.6.3 - Rev: B model:CmpareAEDsys.mdl run by: Ionio solutionMode= STEADY_STATE converge= 1 case: 0 time: 0.000 timeStep:0.0500 therm-package: Janaf Mode:DESIGN iter /pas/Jac/Broy = 5/7/1/3 run: 11/22/06 21:56:58												
Summary Output Data												
MN	Station input	alt (lbm/s)	dTs	W (psia)	Fg (R)	Fn (psia)	T (R)	Tsfc (R)	Wfuel Mach	WAR (ft/s)	OPR (ft ²)	I (lbf)
0.010	FL 0	1500.00	1.40023	14.697	518.68	14.696	518.67	518.67	0.0100	11.17	1756.682	3718053.18
	FL 1	1500.00	1.40023	14.624	518.68	12.327	493.96	493.96	0.5000	544.86	40.871	97952.09
	FL 1.2	1333.33	1.40023	14.624	518.68	12.327	493.96	493.96	0.5000	544.86	36.330	87068.52
	FL 1.3	1333.33	1.39884	22.813	598.23	19.233	569.79	569.79	0.5000	584.96	25.019	93531.76
	FL 1.7	1333.33	1.39884	22.813	598.23	19.233	569.79	569.79	0.5000	584.96	25.019	93531.76
	FL 1.8	1333.33	1.39884	22.356	598.23	14.696	530.71	530.71	0.7979	901.15	19.796	79237.46
	FL 2	166.67	1.40023	14.624	518.68	12.327	493.96	493.96	0.5000	544.86	4.541	10883.57
	FL 2.5	166.67	1.39884	22.813	598.23	19.233	569.79	569.79	0.5000	584.96	3.127	11691.47
	FL 3	166.67	1.35485	380.212	1419.42	357.661	1396.84	1396.84	0.3015	543.67	0.444	25661.88
	FL 3.1	148.33	1.35485	380.212	1419.42	357.661	1396.84	1396.84	0.3015	543.67	0.395	22839.07
	FL 4	152.08	1.28191	365.004	2899.96	0.000	0.00	0.00	0.0000	0.00	0.000	0.00
	FL 4.1	160.42	1.28465	365.004	2830.26	0.000	0.00	0.00	0.0000	0.00	0.000	0.00
	FL 4.4	160.42	1.30687	86.565	2113.04	0.000	0.00	0.00	0.0000	0.00	0.000	0.00
	FL 4.5	168.75	1.30880	86.565	2081.19	0.000	0.00	0.00	0.0000	0.00	0.000	0.00
	FL 7	168.75	1.33625	17.573	1473.06	13.902	1388.26	1388.26	0.6000	1072.49	5.820	17276.39
	FL 8	168.75	1.33625	17.309	1473.12	14.696	1413.45	1413.45	0.4995	900.35	6.677	18853.06
BURNERS	TtOut	2900.00	eff	dPqP	LHV	Wfuel	FAR					
BRN36			0.9950	0.0400	N/A	3.74891	0.02527					

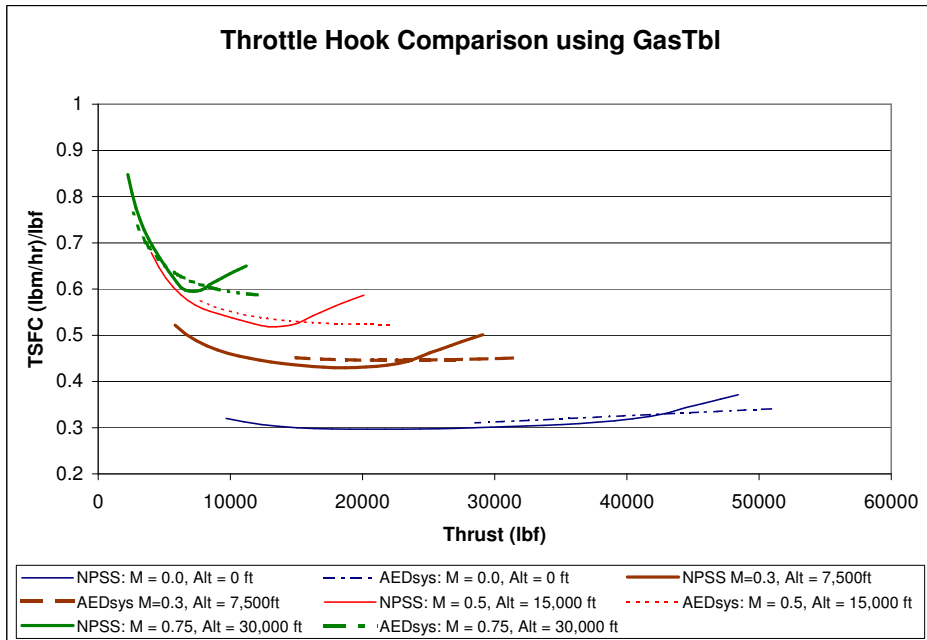


Figure 4.1: Throttle hook comparison of AEDsys and NPSS using the GasTbl thermodynamic package.

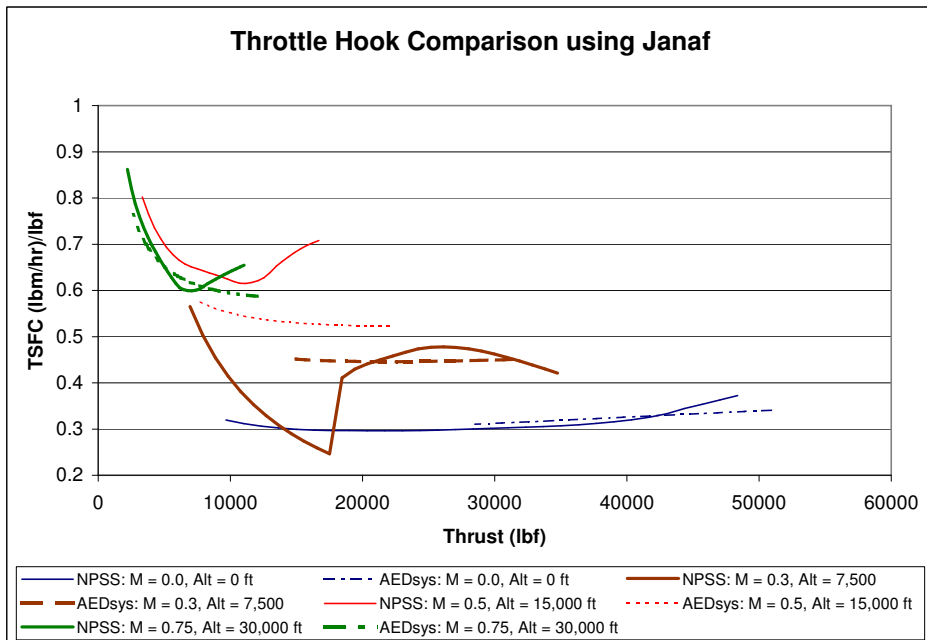


Figure 4.2: Throttle hook comparison of AEDsys and NPSS (using janaf thermodynamics routine) indicating model convergence problems.

difficulty in this implementation. It is unknown why the NPSS solver converged on the solutions at 7,500 and 15,000 feet that are so different than the data generated with an identical model using the GasTbl package. One explanation may be that there are subtle differences in the required inputs for the different thermodynamic packages that were not addressed. Further help from the NASA/GRC group should clear up any remaining difficulties and render the two charts nearly identical.

4.1.3 Explanation of Differences Between AEDsys and NPSS. The two programs display good agreement at the design point, and the off-design performance near the design rpm of the spool also matches closely. Differences for on- and off-design conditions are both noticeable and traceable. A detailed examination of the discrepancies between the two programs' solutions follows.

4.1.3.1 On-design. As seen in tables 4.1 and 4.2, there is an understandable 1% difference between the two net thrust calculations at design conditions. Part of the difference is due to a variation in temperature calculation across the compressor and turbine components as discussed in sections 3.2. The rest of the difference can be calculated by using the specific heat and enthalpy model variations as discussed in section 3.2.1.

The component reports displayed in tables 4.3 and 4.4 show the gradual divergence of total temperature as calculated by the two programs. In order to isolate the difference in power due to variations in total temperature solutions generated by each program we use the T_t and FAR values in the component analysis and calculate enthalpy using only one subroutine. As seen in Table 4.5, the NPSS engine cycle has more power being done across the components. The difference in engine thrust could be balanced by applying the average power difference to the respective spool: approximately 185 to 195 hp on the low pressure spool and 580 to 600 hp on the high pressure spool. The positive number indicates that the power should be applied to the NPSS generated solution, since AEDsys calculated the greater power. However,

Table 4.5: Difference in calculated power due to variations in temperature calculations across components in the NPSS and AEDsys engine cycle programs. For this table, the thermodynamic routine in column 2 was used to define the T_t and FAR while unFAIR was used to calculate the enthalpy.

Component	Routine for T_t	$T_{t\ in} (^{\circ}R)$ $T_{t\ out} (^{\circ}R)$	FAR $\dot{m}_{fuel} (\frac{lbm}{s})$	$h_{in} (\frac{BTU}{lbm})$ $h_{out} (\frac{BTU}{lbm})$	$\dot{m} (\frac{lbm}{s})$	$-\dot{W} (hp)$	AEDsys -NPSS
LPC /FAN	FAIR	518.68	0.00	123.9208	1500	40619.6	
		598.56	0.00	143.0604			
	NPSS (GasTbl)	518.68	0.00	123.9208	1500	40466.82	152.7812
		598.26	0.00	142.9884			
NPSS (janaf)	518.68	0.00	123.9208	1500	40451.54	168.0592	
	598.23	0.00	142.9812				
HPC	FAIR	598.56	0.00	143.0604	166.67	48669.5	
		1425.49	0.00	349.4501			
	NPSS (GasTbl)	598.26	0.00	142.9884	166.67	48326.64	342.8601
		1419.66	0.00	347.9242			
NPSS (janaf)	598.23	0.00	142.9812	166.67	48313.53	355.9703	
	1419.42	0.00	347.8614				
HPT	FAIR	2829.75	0.023886	765.7417	160.41	-49443.2	
		2099.67	3.7421	547.8886			
	NPSS (GasTbl)	2830.35	0.023921	765.9601	160.41	-48628	-815.193
		2112.67	3.74753	551.6988			
NPSS (janaf)	2830.26	0.023929	765.9409	160.42	-48601	-842.18	
	2113.04	3.74891	551.8118				
LPT	FAIR	2068.49	0.02268	538.069	168.74	-41152.4	
		1456.12	3.742136	5.6976			
	NPSS (GasTbl)	2080.72	0.022712	541.6363	168.75	-40936.4	-215.918
		1472.54	3.74753	370.1795			
NPSS (janaf)	2081.19	0.022721	541.7784	168.75	-40931.8	-220.518	
	1473.12	3.74891	370.3408				

this will not resolve the differences because there is another variation between the programs causing the numbers to stand apart.

Following the same analysis described in section 3.2.1 we can calculate the difference in power computed across the compressors and turbines. Both the T_t and FAR values were held constant across all thermodynamic routines for this effort in order to isolate the error due only to variation in specific heat models. As seen in 4.6, the power difference generated by variations of the specific heat/enthalpy model is smaller and in the opposite direction of the temperature induced difference. The average power difference for the low and high pressure spools are -(42 to 52) hp and -(207 to 215) hp respectively. The negative sign indicates that AEDsys has under-

Table 4.6: Difference in calculated power based on variations between specific heat and enthalpy models used for AEDsys and NPSS. Temperatures and FAR values were generated using a rough mean of all solutions, while the enthalpies are calculated using the thermodynamic routine shown in column 2.

Component	Thermo routine	$T_{t\ in} (^{\circ}R)$ $T_{t\ out} (^{\circ}R)$	FAR $\dot{m}_{fuel} (\frac{lbm}{s})$	$h_{in} (\frac{BTU}{lbm})$ $h_{out} (\frac{BTU}{lbm})$	$\dot{m} (\frac{lbm}{s})$	$-\dot{W} (hp)$	FAIR -NPSS
LPC /FAN	FAIR	518.68	0.00	123.9208	1500	41352.94	
		600.00	0.00	143.4059			
	NPSS (GasTbl)	518.68	0.00	124.183	1500	41435.39	-82.4525
		600.00	0.00	143.707			
NPSS (janaf)	518.68	0.00	-6.17665	1500	41444.84	-91.8966	
	600.00	0.00	13.3518				
HPC	FAIR	600.00	0.00	143.4059	166.67	48557.77	
		1425.00	0.00	349.3218			
	NPSS (GasTbl)	600.00	0.00	143.707	166.67	48584.44	-26.6705
		1425.00	0.00	349.736			
NPSS (janaf)	600.00	0.00	13.3518	166.67	48587.08	-29.3116	
	1425.00	0.00	219.392				
HPT	FAIR	2830.00	0.023886	765.8178	160.41	-49438.7	
		2100.00	3.7421	547.9847			
	NPSS (GasTbl)	2830.00	0.023921	767.887	160.41	-49825.8	387.1551
		2100.00	3.74753	548.348			
NPSS (janaf)	2830.00	0.023929	186.544	160.42	-49836.3	397.6378	
	2100.00	3.74891	-33.0275				
LPT	FAIR	2075.00	0.02268	539.9567	168.74	-40375.7	
		1475.00	3.742136	370.8385			
	NPSS (GasTbl)	2075.00	0.022712	540.324	168.75	-40399.0	23.35082
		1475.00	3.74753	371.118			
NPSS (janaf)	2075.00	0.022721	-18.749	168.75	-40369.6	-6.0401	
	1475.00	3.74891	-187.832				

calculated power relative to NPSS based on the specific heat model. That is to say, if the power across the NPSS spools is decreased by the magnitude of these averages it will correct this difference.

When the two errors discussed above are summed, the results is a net power difference between the two programs of 131 to 151 hp on the low pressure spool and 372 to 385 hp on the high pressure spool. The positive number indicates that the AEDsys program has calculated a more power done on each spool, and this can be corrected by applying the power difference to the NPSS shaft. Tables 4.7 and 4.8 show the convergence upon a single solution when the two differences due to the variations in the programs are applied. This analysis only corrected for the differences across

Table 4.7: Comparison of thrust after application of extra shaft Power. Here the HPX is the difference between the two estimates described above.

Program	Thermo Routine	F_n (lbf) uncorrected	% Difference	F_n (lbf) w/ Corrective HPX	% Difference
AEDsys	FAIR	41145			
NPSS	GasTbl	41667	1.268684	41292	0.357
Shaft Power Additions (hp)			LP: 131	HP: 372	

Table 4.8: Comparison of thrust after application of extra shaft power. Here the HPX is the difference between the two estimates described above.

Program	Thermo Routine	F_n (lbf) uncorrected	% Difference	F_n (lbf) w/ Corrective HPX	% Difference
AEDsys	FAIR	41145			
NPSS	janaf	41546	0.974602	41147	0.00486
Shaft Power Additions (hp)			LP: 151	HP: 385	

the turbines and compressors, neglecting any deviations across ducts, mixers, burners, and nozzles. The results for the janaf package presumably are a closer fit because both janaf and FAIR are derived from the same NIST data, so the uncorrected components are not that different from each other. The GasTbl results do accumulate a 0.353% difference across the components.

4.1.3.2 Off-Design. In Figure 4.1 the throttle hooks generated by NPSS and AEDsys mimic each other, sharing a few common points near what would be considered 100% design shaft RPM. The two results diverge as the thrust requirements move the engine away from this central point. This is caused by differences in off-design assumptions. Mattingly [3, p 142] describes the assumptions implemented in AEDsys in detail. The third stated assumption is: off-design component efficiencies do not change from their on-design values. On the other hand, NPSS requires efficiency maps be used for off-design performance analysis. These maps allow the retrieval of an efficiency for the given conditions that may be different than the design condition. The solution for an off-design calculation might iterate several times before converging on a final efficiency number. Compressor and turbine efficiencies corresponding to the data displayed in Figure 4.1 is shown in Figure 4.3 and Figure

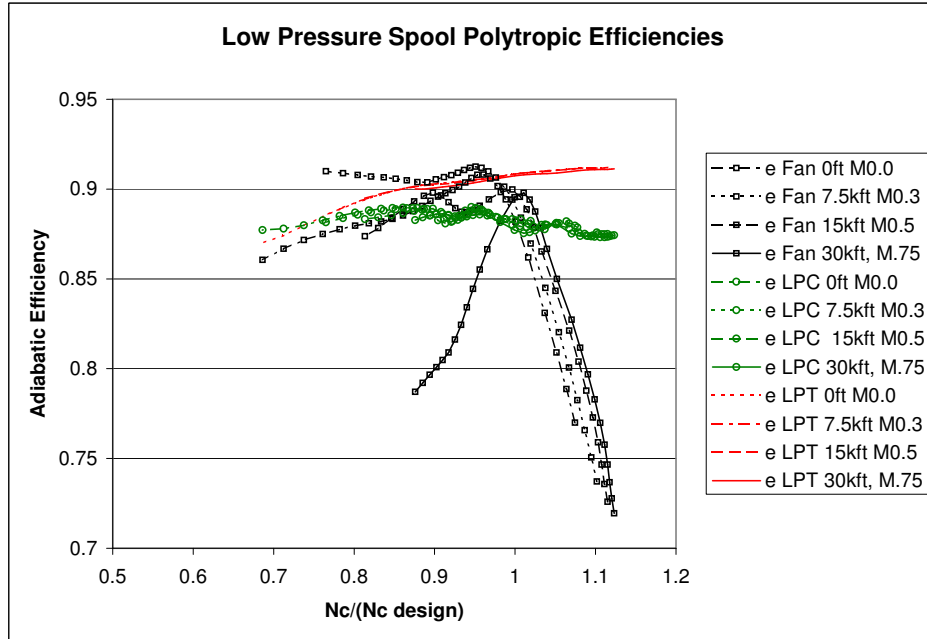


Figure 4.3: Low pressure spool component adiabatic efficiencies returned from the NPSS model using the GasTbl thermodynamic routine.

4.4. Note the severe drop in efficiency experienced by the fan above the 100% design speed. The dramatic change in efficiency is enough to account for the 10% divergence between the two sets of off-design data at high RPM, and the 4% difference at lower RPM.

Another potential source of difference in off-design performance is the implementation of control limits. Limits set in the controls section of AEDsys may be implemented in NPSS, but will affect the performance solution differently due to the differences in thermodynamic routines, off-design assumptions, and efficiency calculations. For the data contributing to Figure 4.1, no limits were imposed upon the NPSS model and the AEDsys model was only constrained by a maximum 110% of on-design rotational speed.

4.2 PDC Hybrid Results

Care was taken to balance the design of the hybrid engine to ensure a legitimate comparison with the baseline high bypass turbofan. Vigilance was required to verify

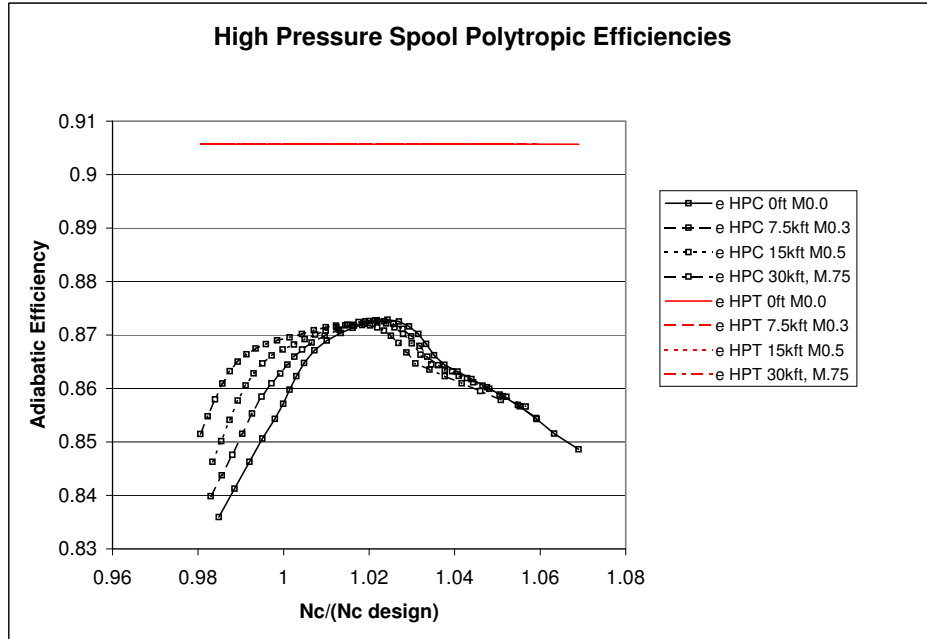


Figure 4.4: High pressure spool adiabatic efficiencies returned by NPSS model using the GasTbl thermodynamics.

the increased benefit apparent in the hybrid results do not stem from a violation of a design limit imposed only on the baseline engine. Most importantly, the total temperature at turbine inlet, and overall pressure ratio were monitored to prevent them from violating the limits for the baseline engine. This is important for on-design performance, but will be crucial in the future off design and mission performance analyses.

4.2.1 Sizing the PDC. Parametric studies were performed to understand the trends associated with the input variables. For the studies in this section we used an initial PDC configuration of 24 tubes as described in Table 4.9. The tubes were run at an 80 + 20 100% fill+purge cycle where the 100% indicates the fill, purge, and fuel masses completely fill the tube. When the results indicated that a 100% purge+fill cycle was not possible the solution was considered invalid and not included in the plots. The results had a few surprises.

Table 4.9: Initial configuration and calculated properties for the pulsed detonation combustor used to generate the trade studies in the results section.

Input Variables	Value	Units
Inner Diameter	2.0	in.
Length	36.0	in
Number of Tubes	24	
AR_{valve}	0.5	
M_{valve}	1.0	
ϕ	0.8992	
pf	0.20	
ff	0.80	
Output Parameter	Value	Units
iBPR	1.45684	
Frequency	59.608	Hz
F_n	27337.6	lbf
TSFC	0.395653	1/hr
T_{t4}	2635.88	R
OPR	38.214	

The parametric studies varied a single design input variable incrementally. The effect on the internal bypass ratio (iBPR), frequency, net thrust (F_n), thrust specific fuel consumption (TSFC), turbine inlet temperature (T_{t4}), and the overall pressure ratio (OPR) were recorded. Experience indicates that the mixer may return a negative Mach number for the secondary inlet flow without returning a program error, so that quantity was included in all output data in order to ensure that a solution was achieved. Also, since the cycle was run at 100%, fill fraction was calculated as $ff = 1.0 - pf$ instead of varying it independently.

4.2.1.1 Tube Diameter. The first thing that needed clarification was how the bypass ratio affected thrust, TSFC, T_{t4} , and overall pressure ratio. It was decided to control this by changing the detonation tube diameter. As seen in Figure 4.5, there is an optimal bypass ratio for the engine when all other parameters are fixed. At this optimal iBPR, the TSFC will hit a minimum. At greater bypass ratio

both thrust and TSFC make unfavorable turns, and below this bypass ratio both the net thrust and the TSFC climb gradually.

There were some issues varying the diameter, and the NPSS code only converged on solutions over a surprisingly small range of diameters. Smaller diameters were limited because the flow through the tube choked. At higher diameters, all the engine core flow was being sucked into the tubes, and none was going into the bypass air, resulting in a hotter turbine inlet temperature and preventing a 100% purge+fill cycle. The range of tube diameters should increase if the tube length is decreased from 36 inches to something shorter.

4.2.1.2 Tube Length. It had been hypothesized that a shorter tube length would allow for increased frequency and greater thrust. However, when we varied tube length and plotted the results seen in Figure 4.6 it was apparent that there would be a benefit to utilizing longer tubes. Not only did thrust increase despite the falling frequency, but there was minimal impact to the total temperature at turbine inlet. If tube length were varied in conjunction with tube diameter to maintain a constant volume, this chart would look significantly different. Perhaps that is something that can be done in the near future.

4.2.1.3 Number of Tubes. Future pulsed detonation combustors will have to deal with the issue of how many tubes to utilize. This has greater ramifications than simply determining iBPR. If sequential firing of some sort is desired, and an effort is made to keep the detonation pulses symmetric about the shafts, it will be advantageous to have a number of tubes easily divisible by $1/\tau_{vo}$ because that term determines how many tubes are required to have a quasi-constant feed from the inlet plenum into the tubes. Also, if firing more than one tube to balance the PDC component, it will help to have an array of tubes divisible by 2, 3, 4, 6, etc.

Number of tubes has the same affect on each of the propulsive properties that the tube diameter did. However, the steps between points are not continuous, since

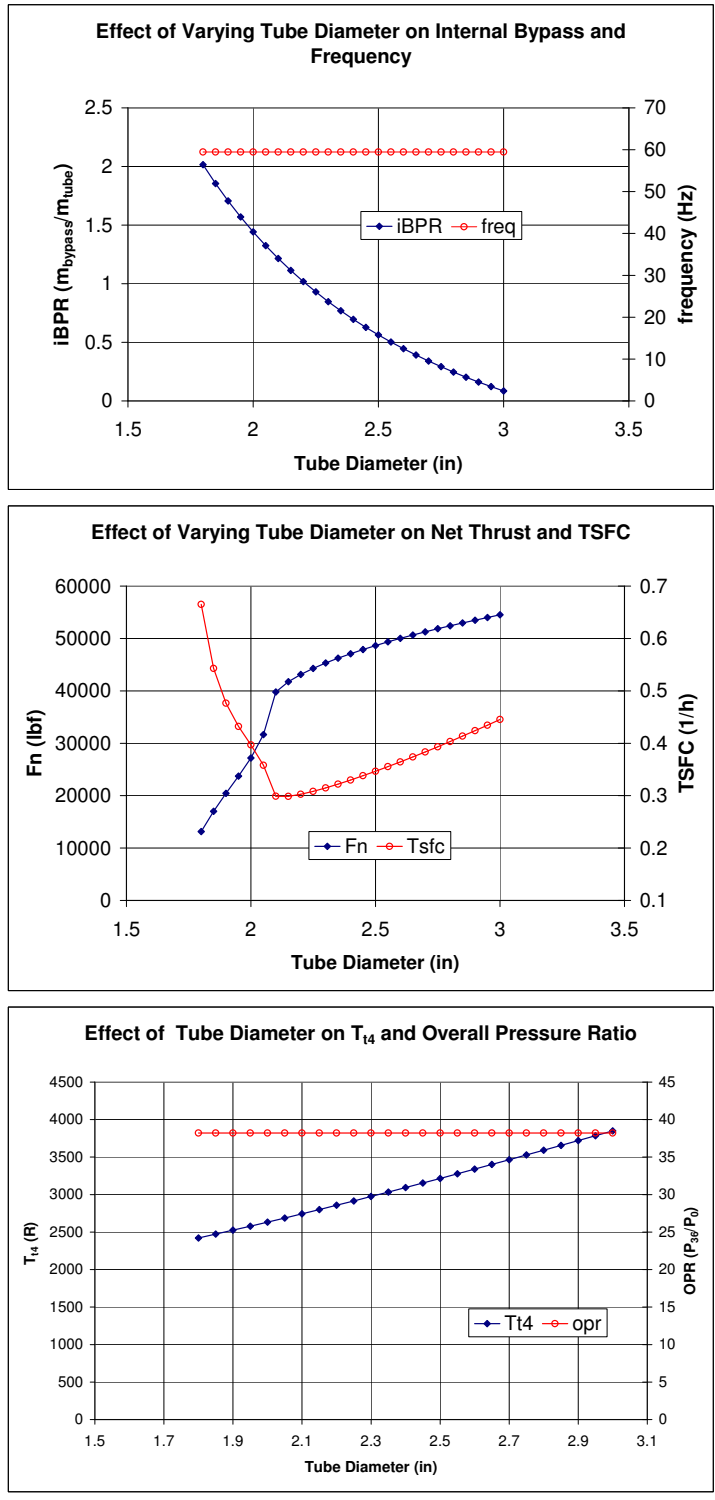


Figure 4.5: Tube diameter is one method of controlling mass flow through the detonation tubes. There is an internal bypass ratio that optimizes the TSFC. Increasing mass flow through the tubes also increases the fuel flow, and raises the temperature of the core flow.

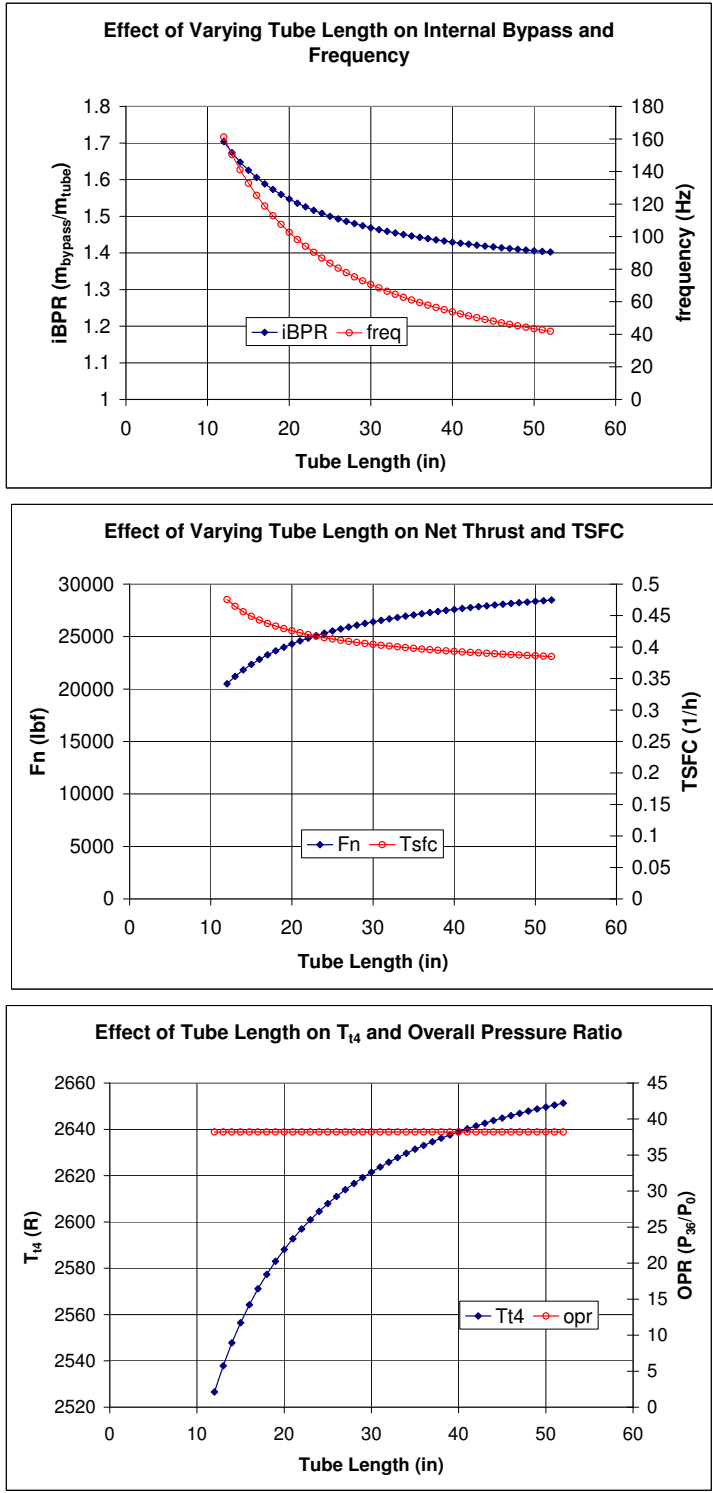


Figure 4.6: Increasing tube length slows the cycle time and allows for greater thrust without a drastic impact on total temperature at turbine inlet.

it is not possible to have $4\frac{1}{2}$ or 13.33333 tubes. It will be necessary to balance the iBPR with a combination of tube diameter and number of tubes.

4.2.1.4 Valve Inlet Area Ratio. One of the most important design choices that could be made is the size of the valves. Mass flow into the tubes is governed by the area of the valve throat. As seen in Figure 4.8, when the area is significantly smaller than the detonation tube diameter, the net thrust and thrust specific fuel consumption will suffer. It should not be assumed that an area ratio of 0.6 is magical simply because the break point in Figure 4.8 is at 0.6; the problem is more complex, and that number might shift if another parameter (such as fill fraction) is modified.

It should be noted that Figure 4.8 indicates that there is a significant benefit to be obtained in this design option, because the turbine inlet temperature does not react drastically, even though the net thrust and TSFC in Figure 4.8 do.

4.2.1.5 Valve Inlet Mach Number. Not surprisingly, the detonation tube inlet valve throat allows better Fn and TSFC for higher Mach numbers. The Mach number and area at the valve throat define the fill time required to fill the tube, which impacts frequency. If geometry is held constant, frequency will define thrust. It appears that most of the benefit of a high filling rate can be obtained by a valve throat Mach number of 0.8 for this configuration. It is anticipated that this trend would continue for other configurations.

4.2.1.6 Compressor Pressure Ratio. Since overall pressure ratio was not changed by any of the previous parametric studies, it was decided to see what effect varying the high pressure compressor had. Figure 4.10 shows the plot. This is the only set of plots that shows a varying OPR, which was expected. It was unexpected to find the minima in TSFC. The TSFC plot of Figure 4.10 shows similar trends to those of Figures 4.5, 4.7, and 4.8, which indicates that the odd shape is due to the changing internal bypass ratio. The baseline engine's high pressure compressor on-

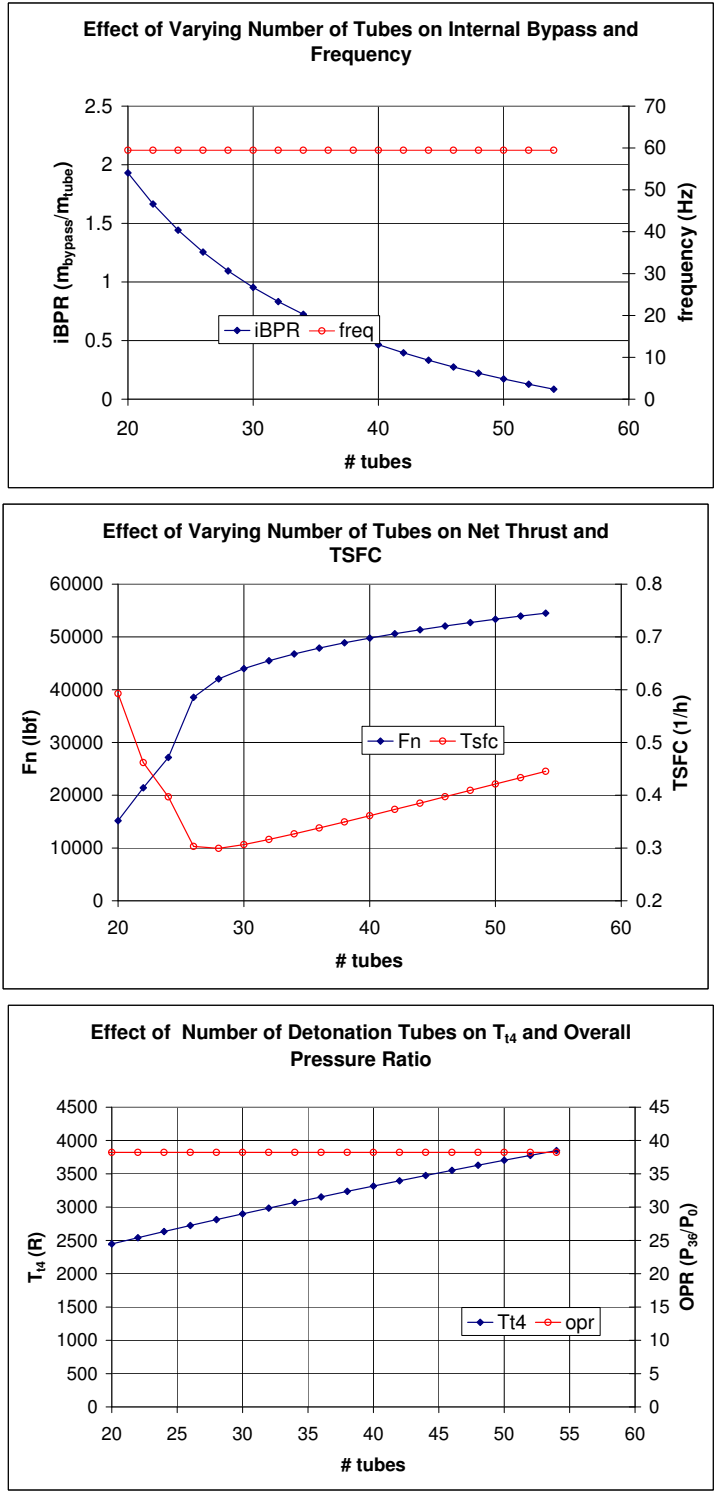


Figure 4.7: Increasing the number of tubes affects the hybrid engine in much the same way as increasing tube diameter. It is the resulting internal bypass that causes the change in performance.

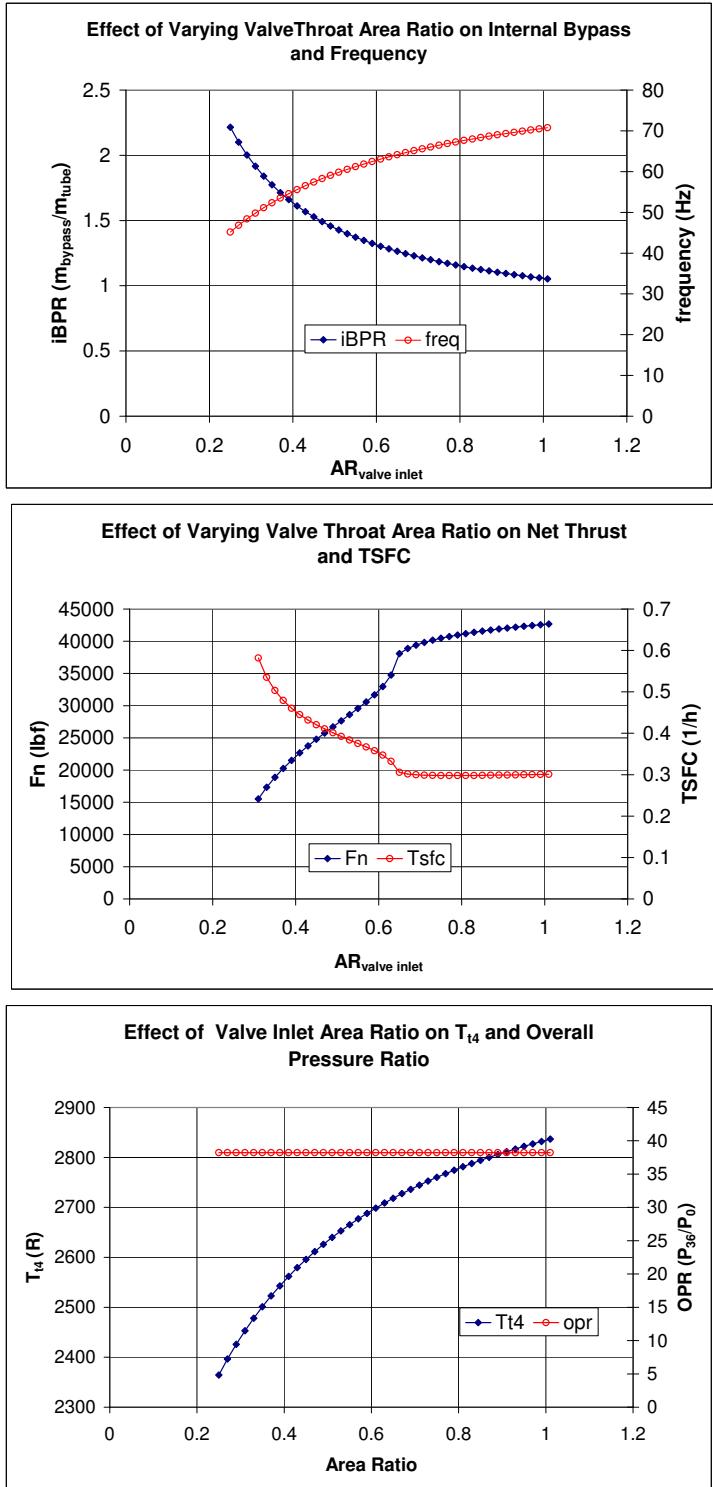


Figure 4.8: Increasing valve inlet area significantly benefits TSFC and F_n while leaving a minimal impact upon T_{t4} .

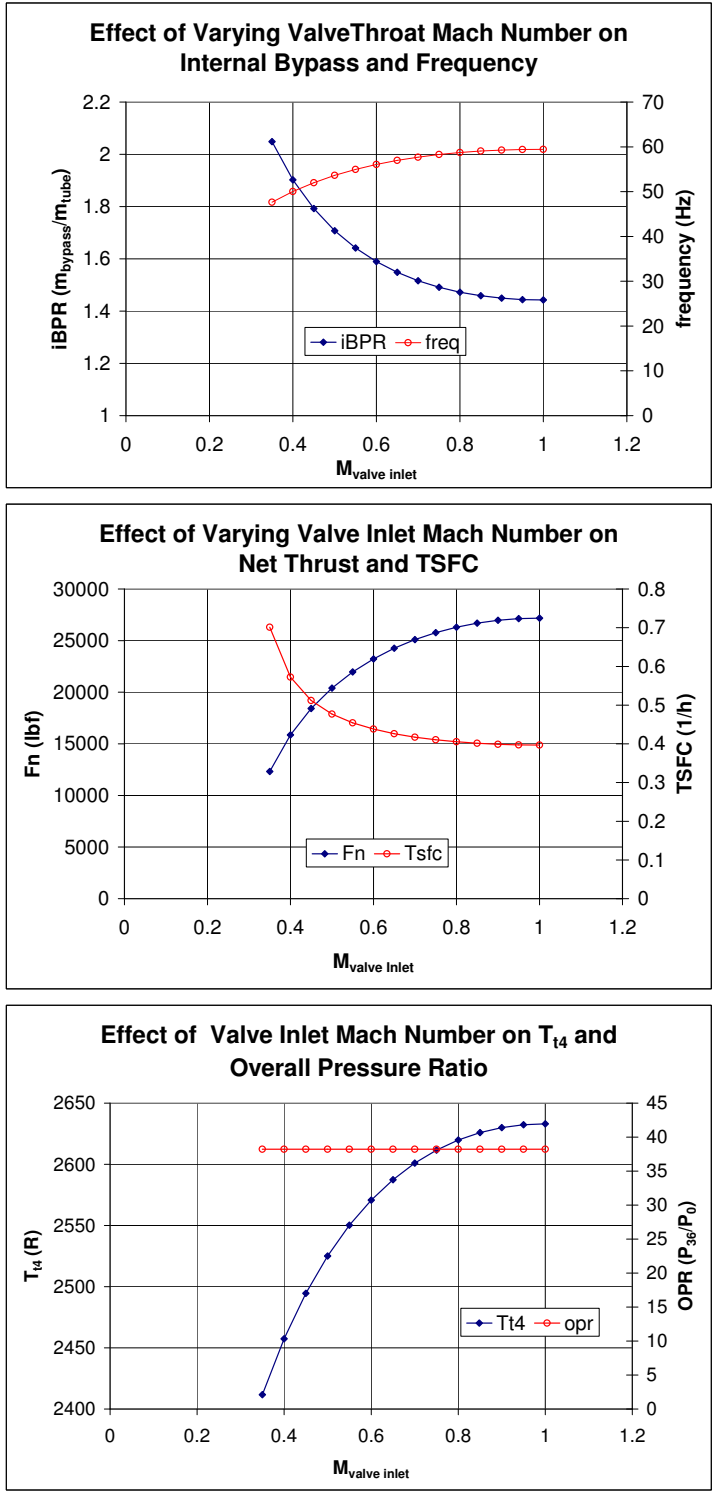


Figure 4.9: Choked flow at valve throat does allow more fill mixture to flow through the tubes (lower iBPR), which causes a rise in T_{t4} . TSFC however, hits a minimum at Mach 0.6 for the selected configuration.

design pressure ratio is 16.66667, which is still on the left-hand portion of the TSFC curve. Increasing the pressure ratio would result in a lower SFC, but the overall pressure ratio is already rather high. The increase in turbine inlet temperature that accompanies the drop in TSFC should not be ignored. It is possible to drop the fuel consumption in other ways without increasing the pressure ratio. For the problem in this thesis, the compressor pressure ratio in the hybrid engine was matched to the ratio in the baseline engine to keep the power across shafts identical.

4.2.1.7 Mixer Primary Inlet Mach Number. Since the NPSS mixer element has proven so difficult, it was decided to see how varying the inlet mach number to the mixer affected the hybrid engine. Varying this parameter had no effect on iBPR, frequency, OPR, or Tt4. However, there was some difference seen in the TSFC and Fn. Figure 4.11 shows the effect that the inlet Mach number has on the engine. Solutions were sought between a Mach number of 0.15 and 1.0, but only achieved above 0.85. At Mach numbers below this value, the solver returned negative values for the secondary inlet flow velocity. Since the mixer uses an energy balance to find the mixed total temperature there is no reason that changing the Mach number at inlet should affect a change in T_{t4} . However, since the mixer routine performs an impulse balance, we do expect to see a change in impulse as inlet Mach numbers vary. This effect should cascade down to the net thrust calculation.

4.2.1.8 Varying Fill, Purge, and Equivalence Ratio in a 100% Cycle. Figures 4.12 and 4.13 show how varying the purge/fill fraction line and equivalence ratio affect the net thrust and TSFC. This has important ramifications for the leap to off-design studies, since this is the anticipated method of thrust control. It should also be noted that the optimal TSFC does not occur at an equivalence ratio of unity.

4.2.1.9 Component Interface Report. A quick search was made based on the results of the parametric studies for a possible hybrid configuration. It was found that a bank of 24 tubes with the characteristics shown in Table 4.12 provided

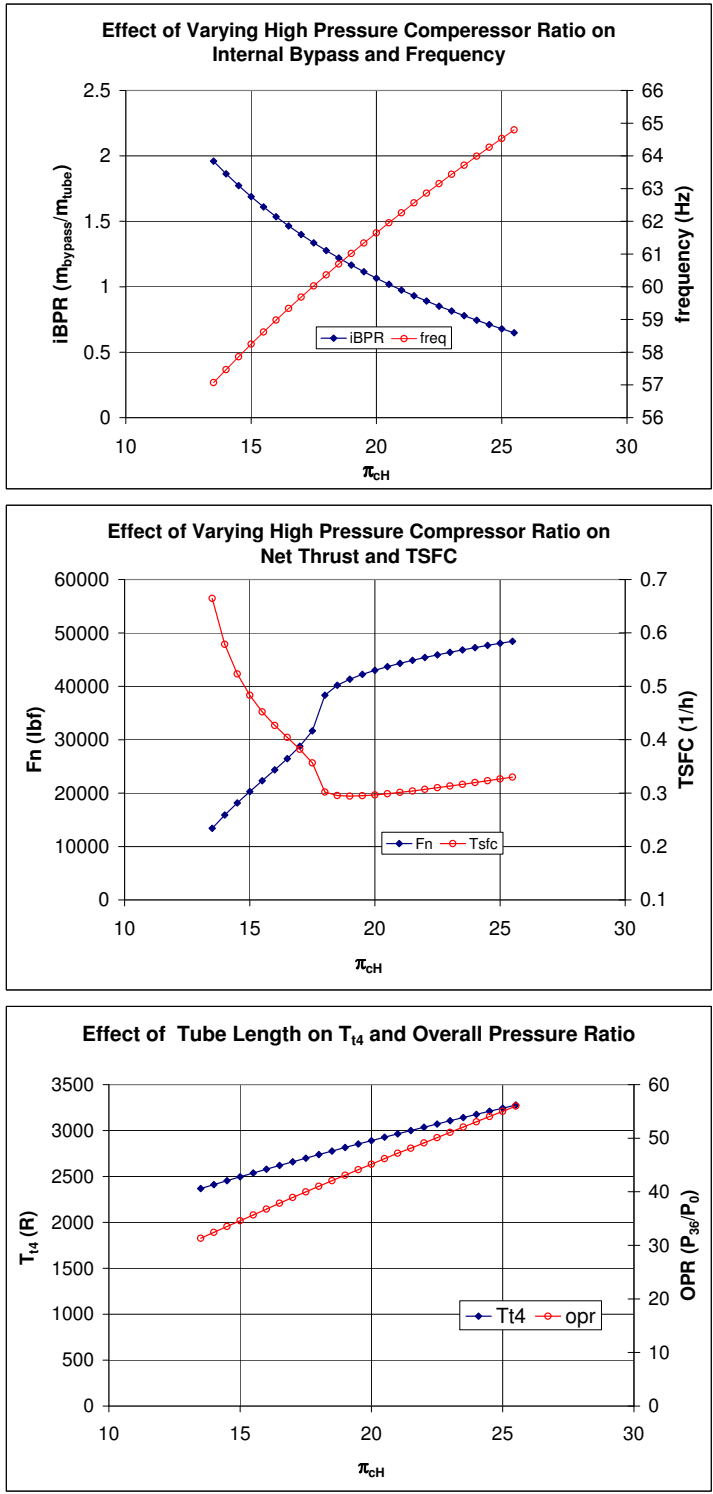


Figure 4.10: Varying the compressor pressure ratio increases T_{t4} while also increasing the frequency and thrust. Internal bypass decreases with increasing compressor ratio.

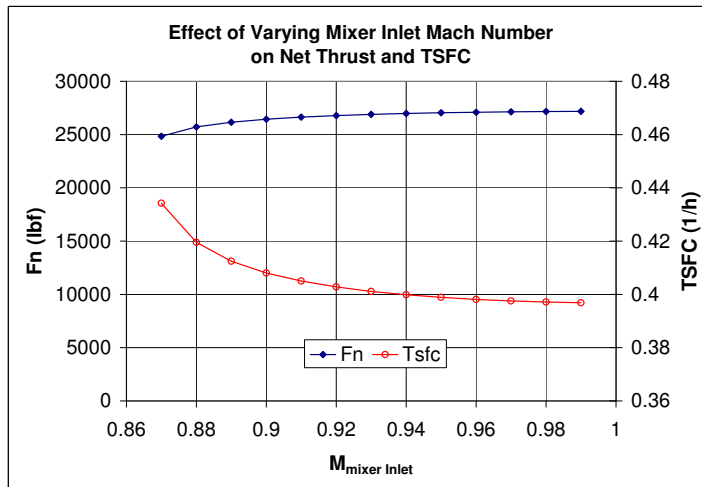


Figure 4.11: Mixer primary inlet Mach number affects only the thrust and TSFC. Presumably this is due to the change in impulse terms generated.

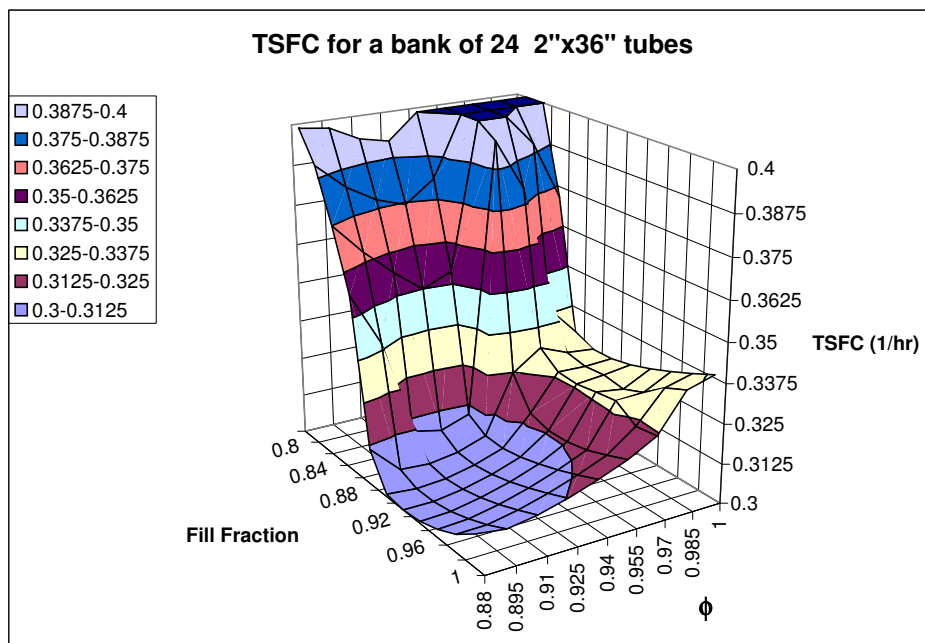


Figure 4.12: Thrust specific fuel consumption is plotted as a function of possible control vectors. For on-design performance, there may be a benefit of utilizing equivalence ratios less than unity.

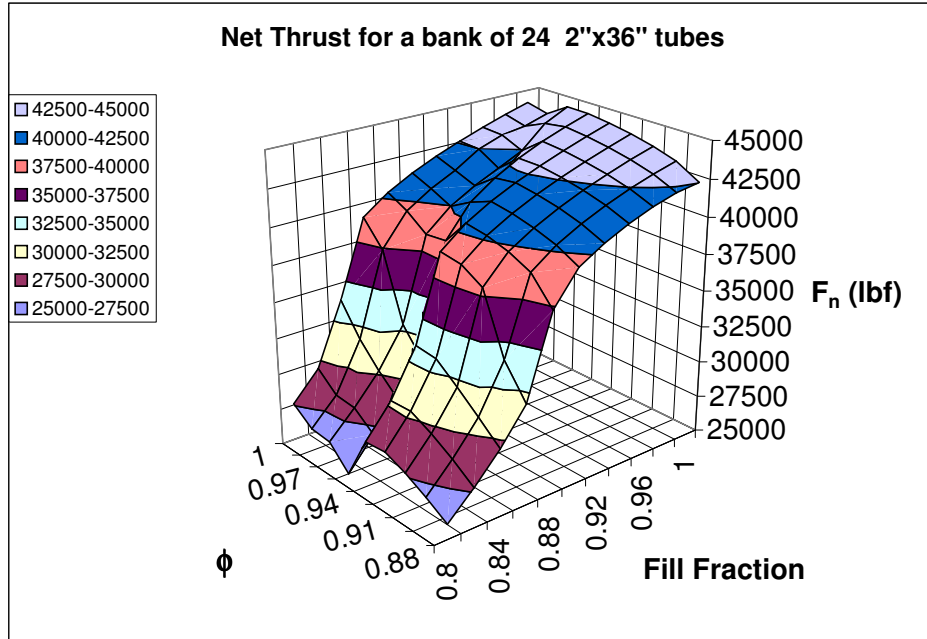


Figure 4.13: Net Thrust plotted as a function of possible control vectors. For on-design performance, Thrust is increased by running with no purge and an equivalence ratio of 1.

an improvement over the baseline engine shown in Table 4.11. This should not be mistaken as an optimized hybrid engine: optimization is a multi-variable non-linear problem requiring more time than is available for this thesis. However, optimization should be done prior to using data to make decisions. The net thrust for the hybrid engine was carefully matched to allow a correct comparison between the two engines. The component interface data is shown in Table 4.12.

4.2.2 Effects of the Transition to Steady State. Shown here is the effect of applying losses through the transition to steady state. For figures 4.14 and 4.15 an array of 24 tubes, with 2.1 inch diameter and 36 inch lengths operating at an 80+20 100% cycle. Area ratio of the detonation tube inlet valves was set at 0.7, with a Mach number of 0.8 was used. Equivalence ratio was set at approximately 90%. It should be compared with the baseline turbofan whose on-design TSFC = 0.3268 1/h and $F_n = 41667$ lbf.

Table 4.10: Improved configuration of the PDC for the hybrid engine based off of the parametric studies.

Parameter	Value	units
Tube Inner Diameter	2.10	inches
Tube Length	36.0	inches
Number of Tubes	24	
AR_{valve}	0.70	
M_{valve}	0.80	
ϕ	0.9012	
pf	0.2349	
ff	0.7651	

Output Parameter	Value	Units
iBPR	1.05101	
Frequency	64.906	Hz
F_n	41677.6	lbf
TSFC	0.298583	1/hr
T_{t4}	2798.11	R
OPR	38.02	

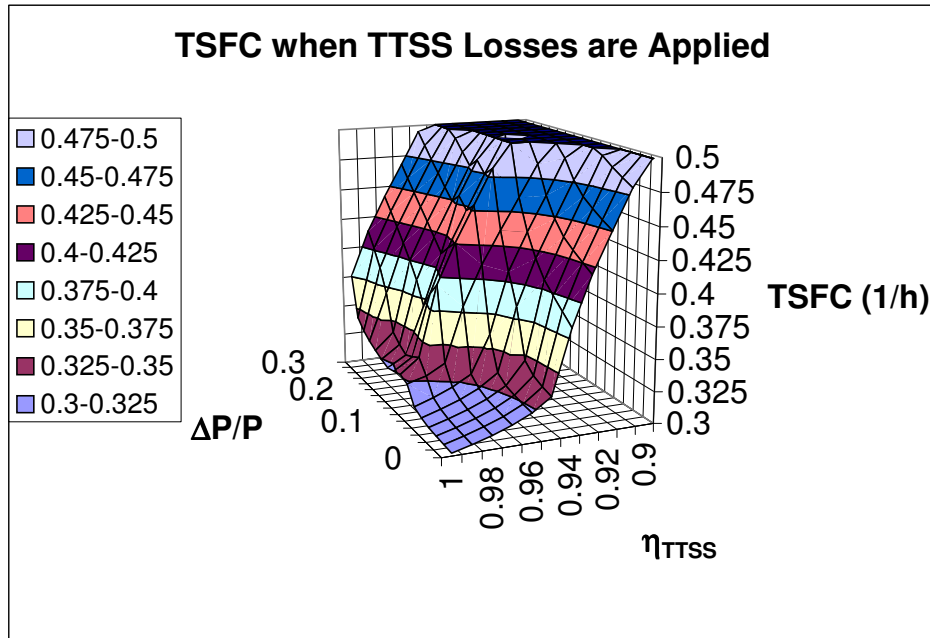


Figure 4.14: Thrust specific fuel consumption is plotted as a function of combustion temperature retention (efficiency) and pressure loss as the flow exiting the PDC transitions from pulsed to steady state.

Table 4.11: NPSS component interface data using the GasTbl thermodynamic package.

NCP NPSS.1.6.3 - Rev: B model:Baseline HBTF bl run by: Ionio solutionMode= STEADY_STATE converge= 1 case: 0 time: 0.000 timeStep:0.0500 therm-package:GasTbl Mode: DESIGN iter /pas/Jac/Broy = 5/7/1/3 run: 02/26/0718 : 01 : 01												
Summary Output Data												
MN	alt	dTs	W	Fg	Fn	TSFC	Wfuel	WAR	OPR			
Station input	\dot{m} (lbm/s)	γ	Pt (psia)	Tt (R)	P (psia)	T (R)	Mach	Velocity (ft/s)	Area (ft ²)	I (lbf)		
FL 0	1500.00	1.39978	14.697	518.68	14.696	518.67	0.0100	11.16	1757.394	3719559.18		
FL 1	1500.00	1.39978	14.623	518.68	12.329	493.97	0.5000	544.91	40.864	97950.38		
FL 12	1333.33	1.39978	14.623	518.68	12.329	493.97	0.5000	544.91	36.324	87067.00		
FL 13	1333.33	1.39963	22.812	598.26	19.286	569.80	0.5000	585.02	24.947	93528.29		
FL 17	1333.33	1.39963	22.812	598.26	19.286	569.80	0.5000	585.02	24.947	93528.29		
FL 18	1333.33	1.39963	22.356	598.26	14.696	530.38	0.8003	903.41	19.734	79200.91		
FL 2	166.67	1.39978	14.623	518.68	12.329	493.97	0.5000	544.91	4.540	10883.38		
FL 2.5	166.67	1.39963	22.812	598.26	19.286	569.80	0.5000	585.02	3.118	11691.04		
FL 3	166.67	1.35513	380.206	1419.66	357.634	1397.08	0.3015	543.78	0.444	25662.10		
FL 3.1	148.33	1.35513	380.206	1419.66	357.634	1397.08	0.3015	543.78	0.395	22839.27		
FL 4	152.08	1.28210	364.998	2900.00	0.000	0.00	0.0000	0.00	0.000	0.00		
FL 4.1	160.41	1.28490	364.998	2830.35	0.000	0.00	0.0000	0.00	0.000	0.00		
FL 4.4	160.41	1.30753	86.667	2112.67	0.000	0.00	0.0000	0.00	0.000	0.00		
FL 4.5	168.75	1.30942	86.667	2080.72	0.000	0.00	0.0000	0.00	0.000	0.00		
FL 7	168.75	1.33651	17.606	1472.54	13.927	1387.77	0.6000	1072.94	5.810	17280.40		
BL 3	1.67	1.35513	380.206	1419.66	0.000	0.00	0.0000	0.00	0.000	0.00		
FL 8	168.75	1.33651	17.342	1472.54	14.696	1412.26	0.5022	905.45	6.640	18801.27		
BURNERS BRN36	TtOut 2900.00	eff 0.9950	dPqP 0.0400	LHV 18400	Wfuel 3.74753	FAR 0.02526						

Table 4.12: NPSS component interface data, for the hybrid engine at design conditions using GasTbl.

NCP NPSS.1.6.3 - Rev: B model: PDC HBTf i4 run by: Ionio solutionMode= STEADY_STATE converge= 1 case: 0 time: 0.000 timeStep:0.0500 therm_package:GasTbl Mode: DESIGN iter/pas/Jac/Broy = 6/8/1/4 run: 02/26/0718 : 11 : 14											
Summary Output Data											
MN	alt	dTs	W	Fg	Fn	TSFC	Wfuel	WAR	OPR	Station input	
0.010	0.0	0.00	1500.0	42189.4	41668.9	0.2978	12406.98	0.0000	38.020		
	\dot{m} (lbm/s)	γ	Pt (psia)	Tt (R)	P (psia)	T (R)	Mach	Velocity (ft/s)	Area (ft ²)	I (lbf)	
FL 0	1500.00	1.39978	14.697	518.68	14.696	518.67	0.0100	11.16	1757.394	3719559.18	
FL 1	1500.00	1.39978	14.623	518.68	12.329	493.97	0.5000	544.91	40.864	97950.38	
FL 12	1333.33	1.39978	14.623	518.68	12.329	493.97	0.5000	544.91	36.324	87067.00	
FL 13	1333.33	1.39963	22.812	598.26	19.286	569.80	0.5000	585.02	24.947	93528.29	
FL 17	1333.33	1.39963	22.812	598.26	19.286	569.80	0.5000	585.02	24.947	93528.29	
FL 18	1333.33	1.39963	22.356	598.26	14.696	530.38	0.8003	903.41	19.734	79200.91	
FL 2	166.67	1.39978	14.623	518.68	12.329	493.97	0.5000	544.91	4.540	10883.38	
FL 2.5	166.67	1.39963	22.812	598.26	19.286	569.80	0.5000	585.02	3.118	11691.04	
FL 3	166.67	1.35513	380.206	1419.66	322.038	1359.00	0.5000	890.15	0.293	18186.40	
FL 3.1	148.33	1.35513	380.206	1419.66	322.038	1359.00	0.5000	890.15	0.261	16185.89	
Fl 3.9	75.78	1.23755	558.776	3948.84	331.327	3552.86	0.9500	2626.45	0.115	11663.06	
Fl 3.92	76.01	1.35513	364.998	1419.66	331.338	1384.05	0.3801	682.50	0.172	9836.24	
FL 4	151.79	1.28625	443.527	2798.11	350.626	2654.71	0.6117	1483.89	0.287	21499.31	
FL 4.1	160.12	1.28892	443.527	2732.72	255.877	2411.89	0.9500	2203.67	0.254	20323.79	
FL 4.4	160.12	1.31257	98.409	2005.42	53.265	1729.44	1.0000	1983.06	0.972	17324.77	
FL 4.5	168.46	1.31430	98.409	1978.18	53.229	1704.62	1.0000	1970.10	1.015	18096.57	
FL 7	168.46	1.34422	17.883	1360.18	14.130	1280.20	0.6000	1033.46	5.475	16551.60	
BL 3	1.67	1.35513	380.206	1419.66	0.000	0.00	0.0000	0.00	0.000	0.00	
FL 8	168.46	1.34422	17.615	1360.18	14.696	1298.37	0.5243	909.04	6.070	17604.71	
BURNERS	TtOut	eff	dPqP	LHV	Wfuel	FAR					
BRN36	4443.75	0.9950	0.0400	18400	3.44638	0.06161					

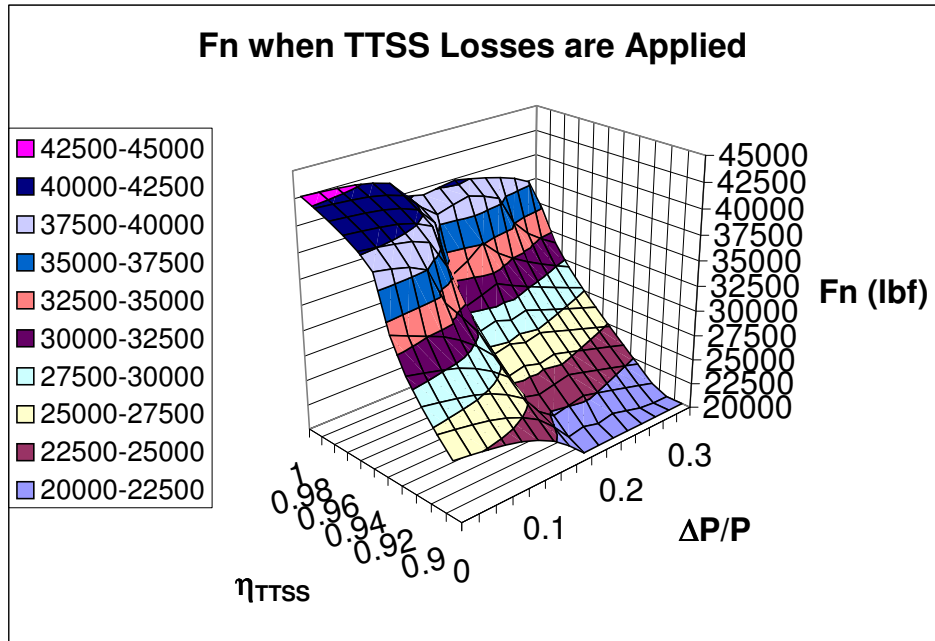


Figure 4.15: Net thrust is plotted as a function of combustion temperature retention (efficiency) and pressure loss as the flow exiting the PDC transitions from pulsed to steady state.

4.2.2.1 *Effects on Thrust.* Thrust is shown in Figure 4.15 to aid the understanding of the TSFC plot. Turbofan engines do not enjoy a constant TSFC throughout their operating range, and often the TSFC decreases with increasing thrust. This said, the peak of the Fn carpet is slightly higher than our baseline engine. All of the thrust between an efficiency of 0.96 and $\frac{\Delta P}{P}$ of 0.3 are above the baseline engine on-design thrust of 41667 lbf.

4.2.2.2 *Effects on TSFC.* Thrust specific fuel consumption (TSFC) for the hybrid engine did see an improvement over the baseline engine. Figure 4.14 shows that the transition-to-steady-state can go through a normalized pressure drop of 30% and still match the baseline engine performance for the chosen configuration. Conversely, if no pressure were lost, it would need to retain 96% of its total temperature rise in order to meet that same level of performance. The plot is marginally optimistic since the Fn values corresponding to the TSFC below 0.325 are above the baseline engine's thrust. By adjusting the $ff + pf$ it is possible to balance the hybrid

engine's thrust with the baseline engine's thrust. This would not significantly change the region of improved TSFC over what is shown here, however.

V. Conclusions

Hybrid turbofan engines utilizing a pulsed detonation combustor continue to hold the promise of improving propulsive efficiency. In order to achieve an improvement in efficiency, total fluid losses before passing through the turbine must be minimized.

5.1 *Baseline Engine Comparison*

In order to make a correct comparison between the hybrid and baseline engine, the baseline engine first had to be shown as acceptable. Comparison efforts demonstrated the agreement of the NPSS model with an industry accepted model in AEDsys. The discrepancies exist due to differences in solution approach, thermodynamic models, and underlying efficiency assumptions.

5.1.1 On-Design Comparison. On-design net thrust and TSFC for the baseline high bypass turbofan engine cycle was shown to agree to within 1.2% between AEDsys and NPSS. When differences in temperature calculations and thermodynamic models are translated into shaft power, the model agreement between the two engines drops to under 0.35%.

5.1.2 Off-Design Comparison. For off-design conditions both programs produce the same trends for F_n and TSFC, but the solutions diverge as the conditions move the shaft RPM from the design point. Close evaluation shows that this divergence is due to Mattingly's [3] assumption of constant efficiency for compressors and turbines. Great care must be taken when utilizing the NPSS package to ensure that the solutions generated are valid.

5.2 *Hybrid Engine Performance*

Only a well-designed PDC will produce a more efficient engine. If minimal losses are experienced in the transition to steady state flow the hybrid engine will outperform the baseline engine.

5.2.1 Geometric Constraints. Parametric studies for the initial configuration indicated that it is important to keep valve openings relatively large. It is also important to optimize the internal bypass ratio by carefully selecting the number and diameter of detonation tubes. Our optimized hybrid engine held 24 tubes that were each 36 inches long and had an inner diameter of 2.1 inches. The tubes were run on an 80% fill + 20% purge 100% total air cycle using an equivalence ratio of approximately 0.9. Valve throat area was set at 70% of the tube cross section, and mass flowed through the throat at Mach number 0.8.

5.2.2 Promise of a More Efficient Engine. Once the design of the PDC has been optimized, the hybrid engine can be evaluated for potential improvements. The detonation combustor describe in section 5.2 showed that an ideal transition to steady state flow would allow a 8.0% decrease in TSFC while maintaining thrust at design conditions. An optimization exercise should be able to increase that improvement, and enlarge the region in which the engine can experience losses while transitioning the pulsed flow to steady state.

5.3 Recommendations

Hybrid turbofan engines utilizing pulsed detonation combustors hold the promise of improved fuel efficiency and greater thrust. The work documented in this thesis should be continued and improved. A greater range of engine parameters needs to be evaluated in order to achieve the best engine configuration. Measurements of how promising the technology can be will only be realized once an optimized PDC is identified.

There are things that should be done to improve the model. Cycle time calculations need to be improved to reflect experimental data. This may require the implementation of CFD studies or look-up tables. Effects of heat transfer through the tube walls might be improved, and the effect of the detonation wave on the purge fluid sitting in the tube should also be considered. A specific architecture for a device

that transitions the flow to steady state could be introduced and programmed. These are things that cannot be completed before this work must be evaluated, but which might be performed in the coming months.

Appendix A. NPSS Model Code

This is the code that defines the models in NPSS. It shows which thermodynamic package was being used, and which elements are called. These files would be called through use of a run file that tells NPSS what to do with the calculations, and then changes variables as needed to generate data.

Listing A.1: Model file for the baseline high-bypass turbofan engine.

```
//
//-----
//  T U R B O J E T   E N G I N E   B U I L D           |
//  |
5 //  B U I L D   A N D   V E R I F Y   T U R B O J E T   |
//  |
//  D E S I G N   P O I N T   O N L Y                 |
//  |
//-----
10 //  T U R B O J E T   C O N F I G U R A T I O N
//-----

cout<<"\t-----\n"
    <<"\t Baseline High Bypass Turbfan Built to Match AEDsys \n"
15    <<"\t-----\n\n"...
    ;

// Set model name
MODELNAME = "Baseline HBTF CmpareAEDsys.mdl with mixers";
//-----
20 // set the thermo package
//-----
//  setThermoPackage("GasTbl");
    setThermoPackage("Janaf");

25 //-----
//  include the standard interpreted things
//-----
#include <InterpIncludes.ncp>
#include "ncp.view"
30 //#include "bleed_macros.fnc"
    //#include "NewDuct.int"

//-----
//  #include the definition file for the user defined engine
35 //  performance component
//-----
#include "EngPerf.cmp" ;

40 //-----
//  MODEL DEFINITION
```

```

//-----
// ##### FLIGHT CONDITIONS #####
Element FlightConditions AMBIENT {
45 // Specify Design conditions
    alt = 0.0; // design altitude (ft)
    MN = 0.01; // design Mach number
    // Ps = 14.696; // ambient pressure (psia)
    // Ts = 59.0; // ambient temperature (F)
50 W = 1500.00; // design mass flow (lbm/s)
}

//##### Inlet #####
Element Inlet INLET {
55 eRamBase = 0.995; //Ram Recovery Factor?
}

// ##### Splitter #####
60 Element Splitter SPLIT {
    BPR = 8.0; // Bypass Ratio
}

65 // ##### FAN #####
// here the fan represents the outer portion of the
// Low pressure compressor spool
Element Compressor Fan21 {
// // use these lines if no compressor map is implemented
70 // effDes = 0.88042; //0.882886;
// PRdes = 1.56;

// use these lines if compressor map is used...
#include "fan.map" ; //Compressor sub-element map
75 S_map.effDes = 0.8827; //0.88289;
S_map.PRdes = 1.56;
}

80 // ##### Bypass Duct/ Nozzle/ Sink #####
Element Duct Bypass13 {
// AEDsys assumes flow in bypass duct is
// isentropic (p109, #9)
}

85 Element Nozzle Noz18 {
    dPqP = 1.0-0.98;
    // pressure loss from nozzle inlet to throat

90 PsExhName = "AMBIENT.Fl_0.Ps";

// AEDsys uses a fixed convergent nozzle
switchType = "CONIC";

```

```

}
95 Element FlowEnd NozSink19 {
}

100 // ##### Low Pressure Compressor #####
Element Compressor LPC20 {
// // use these lines if no compressor map is implemented
// effDes = 0.88042; // set on-x isentropic efficiency
// PRdes = 1.56;
105 // use these lines if compressor map is used...
#include "lpc.map";
S_map.effDes = 0.8827; // set on-x isentropic efficiency
S_map.PRdes = 1.56;
110 }

// ##### High Pressure Compressor #####
Element Compressor HPC25 {
115 // // use these lines if no compressor map is implemented
// effDes = 0.85755; // set on-x isentropic efficiency
// PRdes = 16.66667;

// use these lines if compressor map is used...
120 #include "hpc.map" ; // Compressor sub element map
S_map.effDes = 0.8573 ; // on-x isentropic efficiency
S_map.PRdes = 16.66667 ; // Set on-x pressure ratio

}

125 // ##### Bleed starting point #####
Element Bleed BLD3 {
// ===== BLEEDS =====
// Three Bleeds are taken off of the back side of the
130 // High pressure Compressor
BleedOutPort BL_Cool_301 {
fracW = 0.05; // mass flow (5% for cooling HPT)
}
BleedOutPort BL_Cool_302 {
135 fracW = 0.05; // mass flow (5% for cooling LPT)
}
BleedOutPort BL_Env_303 {
fracW = 0.01; // mass flow fraction (1% bleed)
}
140 }

// ##### Fuel #####
Element FuelStart FUEL32{
145 LHV = 18400; // BTU/lbm - Lower Heating Value of fuel -

```

```

// default is 18400 BTU/lbm

}

150 // ##### Burner #####
Element Burner BRN36{
    effBase = 0.995; // component efficiency
    dPqPBase = 1.0 - 0.96;
    //pi b = 1.0-(dP/P) pressure drop acrss burner
155 // Change from burner default of FAR to TEMPERATURE
    switchBurn = TEMPERATURE;

    // Total temp. at exit (degrees Rankine)
160 //***** not to be used with FAR
    TtCombOut = 2900.0;

}

165 // ##### Bleed Mixer/IGV #####
Element Bleed MIX40 {
    BleedInPort B1In40{
        Pscale = 0.88;
    }
170 }

// ##### HP Turbine #####
Element Turbine HPT41 {
175 #include "hpt.map"; //High Pressure Turbine Map
    S_map.effDes = 0.9057;//0.90555;0.91075;

}

180 // ##### Bleed Mixer #####
Element Bleed MIX44 {
    BleedInPort B1In44{
        Pscale = 0.68;
    }
185 }

// ##### LP Turbine #####
Element Turbine LPT45 {
190 #include "lpt.map" //Low Pressure Turbine Map
    S_map.effDes = 0.9084;//0.90836;0.90906;

}

195 // ##### Nozzle #####
Element Nozzle Noz8 {
    dPqP = 1.0-0.985;
}

```

```

    PsExhName = "AMBIENT.Fl_0.Ps";
    // AEDsys uses a fixed convergent nozzle for core exit
200 switchType = "CONIC";

}

// ##### Terminate Flow #####
205 Element FlowEnd Sink39 {
    // sink for the environmental bleed...
}

Element FlowEnd NozSink9 {
210 // sink for the core airflow
}

// %%%%%%%%%%%
// Put shafts in the model
215 // %%%%%%%%%%%

//##### Low-Pressure Shaft #####
Element Shaft LPShf {
    ShaftInputPort LPC, FAN, LPT ;
220 Nmech = 2000.0; // Shaft RPM at design point
    inertia = 1.0;
    // inertia is only needed for transient analysis

    HPX = 0.0 +131.;
225 // Horsepower extracted from the shaft hp ( = 325.7 kW)

    fracLoss = 1.0 - 0.99;
    // Fractional loss on positive port torque (1.0 - eta_m)

230 }

//##### High Pressure Shaft #####
Element Shaft HPShf {
    ShaftInputPort HPT, HPC ;
235 Nmech = 11000.0;
    inertia = 1.0;

    HPX = 143.178 +372;
    // Horsepower extracted from the shaft hp
240 // ( = 105.7 kW)/ eta m ( = 0.99)

    fracLoss = 1.0 - 0.99;
    // Fractional loss on positive port torque (1.0 - eta_m)

245 }

//##### Engine Performance #####
Element EngPerf PERF{
    // defined in another file to do

```



```

linkPorts("HPC25.Sh_0", "HPShf.HPC", "HP1");
linkPorts("HPT41.Sh_0", "HPShf.HPT", "HP2");

305 // ~~~~~
//   Begin Run Definition
//   ~~~~~

cout << "~~~~~\n"
310 << "           Begin Run Input definitions   \n "
    << "~~~~~\n\n";

```

=====

Listing A.2: Model file for the hybrid high-bypass turbofan engine.

```

//
//-----
//   T U R B O J E T   E N G I N E   B U I L D           |
//   ~~~~~|
5 //   B U I L D   A N D   V E R I F Y   T U R B O J E T   |
//   ~~~~~|
//   D E S I G N   P O I N T   O N L Y                 |
//   ~~~~~|
//-----
10 //   T U R B O J E T   C O N F I G U R A T I O N
//-----

cout << "\t-----\n"
    << "\t Hybrid Pulsed Detonation Combustor High Bypass Turbofan...
        \n"
15 << "\t-----\n\n";

// Set model name
MODELNAME = "PDC HBTF"; // Pulsed Detonation Combustor
                    // High Bypass Turbofan";
20 //-----
// set the thermo package
//-----
    setThermoPackage("GasTb1");
//   setThermoPackage("Janaf");
25 //-----
// include the standard interpreted things
//-----
#include <InterpIncludes.ncp>
30 #include "ncp.view"

//-----
// #include the definition file for the user defined engine
// performance component
35 //-----
#include "EngPerf.cmp" ;

```

```

//-----
40 // MODEL DEFINITION
//-----
// ##### FLIGHT CONDITIONS #####
Element FlightConditions AMBIENT {
    // Specify Design conditions
45 alt = 0.0; // design altitude (ft)
    MN = 0.01; // design Mach number
    W = 1500.00; // design mass flow (lbm/s)
}

50 //##### Inlet #####
Element Inlet INLET {
    eRamBase = 0.995; //Ram Recovery Factor
}

55 // ##### Splitter #####
Element Splitter SPLIT {
    BPR = 8.0; // Bypass Ratio
}

60
// ##### FAN #####
// here the fan represents the outer portion of the Low
// pressure compressor spool
Element Compressor Fan21 {
65 // // use these lines if no compressor map is used
// effDes = 0.88042; //0.882886;
// PRdes = 1.56;

// use these lines if compressor map is used...
70 #include "fan.map" ; //Compressor sub-element map
    S_map.effDes = 0.8827; //0.88289;
    S_map.PRdes = 1.56;
}

75

// ##### Bypass Duct/ Nozzle/ Sink #####
Element Duct Bypass13 {
    // AEDsys assumes flow in bypass duct is
80 // isentropic (p109, #9)
}

Element Nozzle Noz18 {
    // pressure loss from nozzle inlet to throat
85 dPqP = 1.0-0.98;

    PsExhName = "AMBIENT.Fl_0.Ps";
}

```

```

    // AEDsys uses a fixed convergent nozzle for bypass exit
90   switchType = "CONIC";
    }

    Element FlowEnd NozSink19 {

95   }

    // ##### Low Pressure Compressor #####
    Element Compressor LPC20 {
    // // use these lines if no compressor map is used
100  // effDes = 0.88042; // set on-X isentropic efficiency
    // PRdes = 1.56;

    // use these lines if compressor map is used...
105  #include "lpc.map";
    S_map.effDes = 0.8827; //set on-X isentropic efficiency
    S_map.PRdes = 1.56;

    }

110 // ##### High Pressure Compressor #####
    Element Compressor HPC25 {
    // // use these lines if no compressor map is implemented
    // effDes = 0.85755; // set on-X isentropic efficiency
    // PRdes = 16.66667;
115

    // use these lines if compressor map is used...
    #include "hpc.map" ; // Compressor sub element map
    S_map.effDes = 0.8573; // set on-X isentropic efficiency
    S_map.PRdes = 16.66667 ; // Set the pressure ratio on-X
120
    }

    // ##### Bleed starting point #####

125 Element Bleed BLD3 {
    // ===== BLEEDS =====
    // Three Bleeds are taken off of the back side of the
    // High pressure Compressor
    BleedOutPort BL_Cool_301 {
130     fracW = 0.05;
        // mass flow (5% for cooling turbine)
    }
    BleedOutPort BL_Cool_302 {
        fracW = 0.05;
135     // mass flow (5% for cooling turbine)
    }
    BleedOutPort BL_Env_303 {
        fracW = 0.01;
        // mass flow fraction (1% environmental bleed)
140    }

```

```

}

// ##### Fuel #####

145 Element FuelStart FUEL32{

    LHV = 18400; // BTU/lbm - Lower Heating Value of the fuel
                // default is 18400 BTU/lbm

150 }

// ##### Burner #####
#include "PDC_burner.int"
Element PulseDetonationCombustor BRN36{
155     effBase = 0.995;
        dPqPBase = 1.0-0.96;
        switchBurn = FAR;
        FAR = (0.0683 * 1.00); //approx. stoichiometric conditions
160     purgeFrac = 0.2;
        fillFrac = 0.8;
        lTube = 36;
        n_tubes = 24;
        dTube = 2.0;
}
165 // ##### Wall heat exchange #####
//Element Wall WALL38{
// Ahx1 = PI*36; // area of wall inside PDT
// Ahx2 = PI*36*1.02; // area that bypass flow sees
// ChxDes1 = 0.7;//heat transfer film coefficient - blind guess...
170 // ChxDes2 = 0.7;//
// CpMat = 0.1481;//specific heat of material (titanium @ 2160 R)
// // # tubes pi/4 length oD iD(in) rho(lbm/ft...
// ^3) Titanium
// massMat = 36.*(PI/4.*(36./12.)*(2.25**2-2.**2)/144.)*280.93;//...
// mass of material in lbm
//
175 //}

// ##### PDC bypass mixer/ #####
Element Mixer MIX39{
180     Fl_I1.MN = .15;
        Fl_I2.MN = .15;
}

// ##### Bleed Mixer/IGV #####
185 Element Bleed MIX40 {
    BleedInPort B1In40{
        Pscale = 0.88;
    }
190 }

```

```

// ##### HP Turbine #####
Element Turbine HPT41 {
    #include "hpt.map"; //High Pressure Turbine Map
195     S_map.effDes = 0.9057;//0.90555;0.91075;

}
// ##### Bleed Mixer #####
Element Bleed MIX44 {
200     BleedInPort B1In44{
        Pscale = 0.68;
    }

}
205

// ##### LP Turbine #####
Element Turbine LPT45 {
    #include "lpt.map" //Low Pressure Turbine Map
210     S_map.effDes = 0.9084;//0.90836;0.90906;

}

// ##### Nozzle #####
215 Element Nozzle Noz8 {
    dPqP = 1.0-0.985;
    PsExhName = "AMBIENT.Fl_0.Ps";
    switchType = "CONIC";
    // AEDsys uses a fixed convergent nozzle for core exit
220
}

// ##### Terminate Flow #####
Element FlowEnd Sink39 {
225 // BleedInPort B1In44{
//     Pscale = 0.96;
// }
// sink for the environmental bleed...
}
230

Element FlowEnd NozSink9 {
    // sink for the core airflow
}

235 // %%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%
// Put shafts in the model
// %%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%%

//##### Low-Pressure Shaft #####
240 Element Shaft LPShf {
    ShaftInputPort LPC, FAN, LPT ;
    Nmech = 2000.0;

```

```

    inertia = 1.0; // inertia only for transient analysis
245 // Horsepower extracted from the shaft hp ( = 325.7 kW)
    HPX = 0.0;

    // Fractional loss on positive port torque (1.0 - eta_m)
    fracLoss = 1.0 - 0.99;
250 }

//##### High Pressure Shaft #####
Element Shaft HPShf {
    ShaftInputPort HPT, HPC ;
255 Nmech = 11000.0;
    inertia = 1.0;

    // Horsepower extracted from the shaft hp
    // ( = 105.7 kW)/ eta m ( = 0.99)
260 HPX = 143.178;

    // Fractional loss on positive port torque (1.0 - eta_m)
    fracLoss = 1.0 - 0.99;
}
265 //##### Engine Performance #####
Element EngPerf PERF{
}

270 //-----//
//           Flow Connections           //
//                                           //
//           This is where the flow is defined for the engine //
//-----//
275 //

//##### Ambient to Splitter #####
linkPorts( "AMBIENT.Fl_0", "INLET.Fl_I", "FLO" );
linkPorts( "INLET.Fl_0", "SPLIT.Fl_I", "FL1" );
280 //##### Bypass air #####
linkPorts( "SPLIT.Fl_02", "Fan21.Fl_I", "FLb2" );
linkPorts( "Fan21.Fl_0", "Bypass13.Fl_I", "FLb3" );
linkPorts( "Bypass13.Fl_0", "Noz18.Fl_I", "FLb7" );
285 linkPorts( "Noz18.Fl_0", "NozSink19.Fl_I", "FLb8" );

//##### Core Air Flow #####
linkPorts( "SPLIT.Fl_01", "LPC20.Fl_I", "FL2" );
linkPorts( "LPC20.Fl_0", "HPC25.Fl_I", "FL25");
290 linkPorts( "HPC25.Fl_0", "BLD3.Fl_I", "FL3" );
linkPorts( "BLD3.Fl_0", "BRN36.Fl_I", "FL31");
//##### Fuel Flow #####
linkPorts( "FUEL32.Fu_0", "BRN36.Fu_I", "Fu3" );
//linkPorts( "BRN36.Fl_01", "WALL38.Fl_I1", "Wa1" );

```


Appendix B. Fortran unFAIR Code

In order to better understand the differences between AEDsys and NPSS, the underlying thermodynamic routines were evaluated. Since the FAIR routine from AEDsys was not available as a callable routine or stand-alone package, it was coded in FORTRAN based on the formulas included in Mattingly [28]. Output was checked with the tabulated results included in that same text.

Listing B.1: Constants module used for the unFAIR subroutine.

```
module constant_module

! Programmer: 1Lt Ionio Andrus
! AFIT/ENY MENG 732
5 ! 16 Apr 05

integer, public, parameter :: RKIND = selected_real_kind(13)

real(kind = RKIND), public, parameter :: zero = 0.0_RKIND
10 real(kind = RKIND), public, parameter :: one = 1.0_RKIND
real(kind = RKIND), public, parameter :: two = 2.0_RKIND
real(kind = RKIND), public, parameter :: three = 3.0_RKIND
real(kind = RKIND), public, parameter :: four = 4.0_RKIND
real(kind = RKIND), public, parameter :: five = 5.0_RKIND
15 real(kind = RKIND), public, parameter :: six = 6.0_RKIND
real(kind = RKIND), public, parameter :: seven = 7.0_RKIND
real(kind = RKIND), public, parameter :: eight = 8.0_RKIND
real(kind = RKIND), public, parameter :: nine = 9.0_RKIND
real(kind = RKIND), public, parameter :: ten = 10.0_RKIND
20 real(kind = RKIND), public, parameter :: half = 0.5_RKIND
real(kind = RKIND), public, parameter :: quarter = 0.25_RKIND
real(kind = RKIND), public, parameter :: tenth = 0.1_RKIND
real(kind = RKIND), public, parameter :: one_n_half = 1.5_RKIND
real(kind = RKIND), public, parameter :: thousand = 1000.0_RKIND
25 !real(kind = RKIND), public, parameter :: zero = 0.0_RKIND
!real(kind = RKIND), public, parameter :: zero = 0.0_RKIND
!
real(kind = RKIND), public, parameter :: R_Air = 286.96_RKIND !J/...
kgK

30 !define some conversion multipliers
real(kind=RKIND), public, parameter :: K_to_R = 1.800
real(kind=RKIND), public, parameter :: kJ_kg_to_BTU_lbm = 0.4299226139
real(kind=RKIND), public, parameter :: BTU_lbm_to_ft2_s2 = 25037.00
real(kind=RKIND), public, parameter :: BTU_s_to_hp = 1.41485320412
35

end module constant_module
```

Listing B.2: Main callable unFAIR subroutine.

```

subroutine unFAIR(T, h, Pr, phi, Cp, R, gam, a, FAR)
use constant_module
implicit none

5 REAL (kind = RKIND) :: A0, A1, A2, A3, A4, A5, A6, A7
REAL (kind = RKIND) :: h_ref, phi_ref, phi_r1, phi_r2
REAL (kind = RKIND) :: T, FAR !intent in
REAL (kind = RKIND) :: Pr, R, Cp, h, phi, gam, a !intent out
REAL (kind = RKIND) :: cp_a, h_a, phi_a, cp_p, h_p, phi_p
10 character *3 :: flow

!=====
!Check to see if FAR affects
if (FAR .lt. 1.0E-9_RKIND) flow = "no"
15 !===== Define coefficients from Table 2.2 for air alone =====
A0 = 2.5020051E-01_RKIND
A1 = -5.1536879E-05_RKIND
A2 = 6.5519486E-08_RKIND
20 A3 = -6.7178376E-12_RKIND
A4 = -1.5128259E-14_RKIND
A5 = 7.6215767E-18_RKIND
A6 = -1.4526770E-21_RKIND
A7 = 1.0115540E-25_RKIND
25 h_ref = -1.7558886_RKIND !BTU/lbm
phi_r1 = 0.0454323_RKIND !BTU/(lbm R)

!===== Equations 2.60,2.61, 2.62 for air alone =====
cp_a = A0 + A1*T + A2*T**2 + A3*T**3 + &
30 A4*T**4 + A5*T**5 + A6*T**6 + A7*T**7

h_a = h_ref +A0*T +A1/two*T**2 +A2/three*T**3 +A3/four*T**4 +&
A4/five*T**5 + A5/six*T**6 + A6/seven*T**7 + A7/eight*T**8

35 phi_a = phi_r1 +A0*log(T) + A1*T +A2/two*T**2 +A3/three*T**3 +&
A4/four*T**4 + A5/five*T**5 + A6/six*T**6 + A7/seven*T**7
!=====

!==== Now change coefficients for the products of combustion. =
40 A0 = 7.3816638E-02_RKIND
A1 = 1.2258630E-03_RKIND
A2 = -1.3771901E-06_RKIND
A3 = 9.9686793E-10_RKIND
A4 = -4.2051104E-13_RKIND
45 A5 = 1.0212913E-16_RKIND
A6 = -1.3335668E-20_RKIND
A7 = 7.2678710E-25_RKIND
h_ref = 30.58153_RKIND !BTU/lbm
phi_r2 = 0.6483398_RKIND !BTU/(lbm R)
50 !===== Equations 2.60, 2.61, 2.62 for products of combustion =

```

```

cp_p = A0 + A1*T + A2*T**2 + A3*T**3 + &
      A4*T**4 + A5*T**5 + A6*T**6 + A7*T**7

55 h_p = h_ref + A0*T + A1/two*T**2 + A2/three*T**3 + A3/four*T**4 +&
      A4/five*T**5 + A5/six*T**6 + A6/seven*T**7 + A7/eight*T**8

phi_p = phi_r2 + A0*log(T) + A1*T + A2/two*T**2 + A3/three*T**3 +&
      A4/four*T**4 + A5/five*T**5 + A6/six*T**6 + A7/seven*T**7
60 !=====

!=====      Equation 4.26 a,b,c,d      =====
R = 1.9857117_RKIND / (28.97_RKIND - FAR*0.946186_RKIND)
65 !BTU/(lbm R)
Cp = (Cp_a + FAR*Cp_p)/(one+FAR)
h = (h_a + FAR*h_p)/(one + FAR)
phi = (phi_a + FAR*phi_p)/(one + FAR)

70 !=====      Equation 2.55 - "reduced pressure "      =====
phi_ref = 1.578420959_RKIND      ! BTU/(lbm R)  phi@492.00 R
      ! (a hair above freezin')
Pr = exp((phi - phi_ref)/R)

75 !=====      do the additional calculations      =====
gam = Cp/ (Cp-R)
a = sqrt(gam*R*BTU_lbm_to_ft2_s2*T)
!=====

80 !=====
end subroutine

```

Appendix C. Pulsed Detonation Combustor Code

This appendix contains the NPSS code for the pulsed detonation combustor element used by the NPSS model. The code is based on the work burner element generated by NASA/GRC that was included with the NPSS release 1.6.3.

Listing C.1: NPSS element file for the pulsed detonation combustor.

```
#ifndef __PDC__
#define __PDC__

//*****
5 // * Air Force Institute of Technology
// * 2950 Hobson Way, Bldg 641
// * Wright Patterson AFB, OH 45433
// *
// * Ionio Q. Andrus, Capt., USAF
10 // *
// BASED ON "Burner.int" included in NPSS, written by~~
// * NASA Glenn Research Center
// * 21000 Brookpark Rd
// * Cleveland, OH 44135
15 // *
//*****

#include <InterpIncludes.ncp>

20 class PulseDetonationCombustor extends Element {

//-----
//      ***** DOCUMENTATION *****
//-----

25     title = "";

        description = isA() + " will calculate performance for
        pulsed detonation combustor.";

30     usageNotes = "

The burner element performs high level burner performance
calculations. This element works with an entrance fluid and
35 fuel stream. It mixes the two flows together and then
performs the burn calculations. Please note that the burner
has no control over the actual fuel stream conditions--fuel type,
LHV, etc. These values are properties of the fuel flow itself
and are usually set in the FuelStart element.

40 There are two ways to specify the burner exit conditions. The
first way is specify the burner fuel-to-air ratio. The second
way is to set equivalence ratio. The type of input used is
controlled by an option switch.
```

45

The burner tracks several different pressure losses. The first, $dPqP$, accounts for duct friction pressure drops and approximates the pressure loss through valves. The second, $dPqPRayleigh$, accounts for the Rayleigh pressure drop. $dPRayleigh$ is input or
50 calculated - see `switchHotLoss`, an iteration is necessary since the pressure loss itself is a function of the exit conditions.

The burner also allow two efficiencies to be input. The first efficiency, `eff`, refers to the efficiency based on enthalpy
55 change. The second efficiency, `effChem`, refers to the efficiency based on temperature change. Both terms can be input. However, the enthalpy efficiency is always applied first.

Additionally,

60

The user can request a pre burner pressure loss $dPqP$. The pressure loss calculations are performed before all the other calculations are done. This means that the combustion entrance pressure will not match the value indicated by the burner entrance.

65

The user can request a heat transfer Qhx . The heat transfer calculations are performed after all the other calculations are done. This means that if heat transfer is being used, the exit temperature will not match the value indicated by the burner
70 calculations.

";

background = "";

75

```
//-----  
//      ***** SETUP VARIABLES *****  
//-----
```

80

```
real a_dPqP {  
  value = 0.0;  IOstatus = "input";  units = "none";  
  description = "Duct friction pressure drop adder";  
}
```

85

```
real a_dPqPAud {  
  value = 0.0;  IOstatus = "unset";  units = "psia";  
  description = "Audit factor adder applied to pressure ratio";  
}
```

90

```
real a_eff {  
  value = 0.0;  IOstatus = "input";  units = "none";  
  description = "Adiabatic efficiency adder";  
}
```

95

```
real a_effChem {  
  value = 0.0;  IOstatus = "input";  units = "none";  
  description = "Chemical efficiency adder";  
}
```

```

real ARvalve {      // Added 15Feb2007 - IA
  value = 0.5; IOstatus = "input"; units = "none";
  description = "Ratio of valve throat area to tube cross section...
    area";
100 }
real deltaS {      //Added 17Jan2007 - IA
  value = 0.0; IOstatus = "output"; units = "none";
  description = "Change in entropy due to detonation";
}
105 real DDT {      //Added 17Jan2007 - IA
  value = 0.0005; IOstatus = "input"; units = "none"; //...
    seconds
  description = "Detonation to deflation time in seconds";
}
real dPqP {
110  value = 0.0; IOstatus = "output"; units = "none";
  description = "Adjusted duct friction pressure drop";
}
real dPqPBase {
115  value = 0.0; IOstatus = "input"; units = "none";
  description = "Duct friction pressure drop ";
}
real dPqPRayleigh {
  value = 0.0; IOstatus = "input"; units = "none";
  description = "Adjusted Rayleigh pressure drop";
120 }
real dTube {      //Added 17Jan2007 - IA
  value = 2.0; IOstatus = "input"; units = "none"; // inches...
    ...
  description = "Inside diameter of the detonation tube";
}
125 real eff {
  value = 1.0; IOstatus = "output"; units = "none";
  description = "Adjusted adiabatic burner efficiency";
}
real effBase {
130  value = 1.0; IOstatus = "input"; units = "none";
  description = "Adiabatic burner efficiency, from socket ";
}
real effChem {
135  value = 1.0; IOstatus = "input"; units = "none";
  description = "Adjusted chemical efficiency";
}
real effChemBase {
  value = 1.0; IOstatus = "input"; units = "none";
  description = "Chemical efficiency, from socket";
140 }
real eqRatio {
  value = 1.0; IOstatus = "input"; units = "none";
  description = "Equivalence ratio for fuel-air mixture";
}
145 real FAR {

```

```

    value = 0.0; IOstatus = "output"; units = "none";
    description = "Fuel-to-air ratio";
}
real FARDes {
150   value = 0.0; IOstatus = "output"; units = "none";
    description = "Fuel-to-air ratio at design";
}
real fillFrac { //Added 17Jan2007 - IA
155   value = 1.0; IOstatus = "input"; units = "none";
    description = "Fill fraction ";
}
real fuelFractV {
160   value = 0.0; IOstatus = "input"; units = "none";
    description = "Fraction of the incoming flow velocity fuel
enters the burner";
}
real iBPR { //added 17Jan2007 - IA
165   value = 1.0; IOstatus = "output"; units = "none";
    description = "Bypass ratio internal to the PDC";
}
real iBPRdes { //added 1Feb2007 - IA
170   value = 1.0; IOstatus = "output"; units = "none";
    description = "Bypass ratio internal to the PDC at
design conditions";
}
real lTube { //added 17Jan2007 - IA
175   value = 36; IOstatus = "input"; units = "none"; //inches??
    description = "length of the individual detonation tubes";
}
real n_tubes{ //added 17Jan2007 - IA
180   value = 36; IOstatus = "input"; units = "none";
    description = "Total number of detonation tubes used
in the PDC";
}
real MCJ { //added 17Jan2007 - IA
185   value = 3.0; IOstatus = "output"; units = "none";
    description = "Chapman-Jouguet Mach number of the
detonation wave.";
}
real Mvalve { //added 15Feb2007 - IA
190   value = 1.0; IOstatus = "input"; units = "none";
    description = "Mach number of flow passing through
the valve throat.";
}
real qadd{ //added 17Jan 2007- IA
195   value = 0.0; IOstatus = "output"; units = "none";
    description = "Heat addition due to fuel combustion";
}
real Qhx {
    value = 0.0; IOstatus = "input"; units = "Btu/sec";
    description = "Heat loss to thermal mass storage";
}

```

```

real PqPRayleigh {
  value = 1.0; IOstatus = "output"; units = "none";
200   description = "Adjusted Rayleigh pressure drop";
}
real PqPRayleighDelta {
  value = 0.0; IOstatus = "output"; units = "none";
  description = "Bounded Rayleigh pressure drop - for loop only"...
;
205 }
real PqPRayleighError {
  value = 1.0; IOstatus = "output"; units = "none";
  description = "Adjusted Rayleigh pressure drop error";
}
210 real PqPRayleighMin {
  value = 0.05; IOstatus = "input"; units = "none";
  description = "Rayleigh pressure drop lower limit - for loop ...
    only";
}
real PqPRayleighStep {
215   value = 0.05; IOstatus = "input"; units = "none";
  description = "Maximum step for Rayleigh pressure drop
    - for loop only";
}
real PqPRayleighNew {
220   value = 1.0; IOstatus = "output"; units = "none";
  description = "Previous adjusted Rayleigh pressure drop
    - for loop only";
}
real purgeFrac { //Added 17Jan2007 - IA
225   value = 0.25; IOstatus = "input"; units = "none";
  description = "Purge fraction coefficient for flow";
}
real s_dPqP {
  value = 1.0; IOstatus = "input"; units = "none";
230   description = "Duct friction pressure drop scalar";
}
real s_dPqPAud {
  value = 1.0; IOstatus = "unset"; units = "none";
  description = "Audit factor scalar applied to pressure ratio";
235 }
real s_eff {
  value = 1.0; IOstatus = "input"; units = "none";
  description = "Adiabatic efficiency scalar";
}
240 real s_effChem {
  value = 1.0; IOstatus = "input"; units = "none";
  description = "Chemical efficiency scalar";
}
real tauBlDn { // Added 17Jan2007 - IA
245   value = 5.; IOstatus = "input"; units="none";
  description = "Blowdown time constant";
}

```

```

real tauValveOpen { // Added 18Jan2007 - IA
    value = 0.33333; IOstatus = "output"; units="none";
250   description = "time valve open/ time cycle - from 0 to 1";
}
real tCycle{ // Added 17Jan2007 - IA
    value = 0.01; IOstatus = "output"; units = "none"; //seconds
    description = "Detonation engine cycle time (= 1/frequency)"...
    ;
255 }
real tolRayleigh {
    value = 4e-05; IOstatus = "input"; units = "none";
    description = "Iteration tolerance on momentum pressure drop";
}
260 real tolWfuel {
    value = 1e-05; IOstatus = "input"; units = "none";
    description = "Iteration tolerance on temperature burn";
}
real TtCombOut {
265   value = 0.0; IOstatus = "input"; units = "R";
    description = "Exit temperature";
}
real TtLast {
    value = 0.0; IOstatus = "input"; units = "R";
270   description = "Previous exit temperature - for loop only";
}
real TTSSeff{ // Added 17Jan2007 - IA
    value = 1.0; IOstatus = "input"; units = "none";
    description = "Efficiency factor for the transition device."...
    ;
275 }
real TTSSdPqP{ // Added 17Jan2007 - IA
    value = 0.0; IOstatus = "input"; units = "none";
    description = "Change in Pressure divided by Pressure
    for transition to steady state calculation.";
280 }
real tValve{ // Added 17Jan2007 - IA
    value = 0.0002; IOstatus = "input"; units = "none"; //...
    seconds
    description = "Time for valves to open/close";
}
285 real Wfuel {
    value = 0.0; IOstatus = "input"; units = "lbm/sec";
    description = "Combustor fuel flow";
}
real WfuelError {
290   value = 0.0; IOstatus = "input"; units = "lbm/sec";
    description = "Combustor fuel flow error";
}
real WfuelLast {
    value = 0.0; IOstatus = "input"; units = "lbm/sec";
295   description = "Previous combustor fuel flow - for loop only";
}

```

```

real WfuelNew {
    value = 0.0; IOstatus = "input"; units = "lbm/sec";
    description = "Next combustor fuel flow - for loop only";
300 }
int countFuel {
    value = 0; IOstatus = "output";
    description = "Fuel loop counter";
}
305 int countFuelMax {
    value = 50; IOstatus = "input";
    description = "Fuel loop maximum counter";
}
int countRayleigh {
310 value = 0; IOstatus = "output";
    description = "Rayleigh loop counter";
}
int countRayleighMax {
    value = 25; IOstatus = "input";
315 description = "Rayleigh loop maximum counter";
}
int flagRayleighLossTooMuch {
    value = 0; IOstatus = "output";
    description = "If true, Rayleigh loop results in too much loss...
        ";
320 }
int flagRayleighChoked {
    value = 0; IOstatus = "output";
    description = "If true, Rayleigh loop results in supersonic ...
        flow";
}
325 // for backward compatibilty with old "aud"
FunctVariable a_dPqPAud {
    units = "none"; IOstatus = "input";
    getFunction = "get_aAud"; setFunction = "set_aAud";
330 }
real get_aAud() { return a_dPqPAud; }
void set_aAud(real userValue) { a_dPqPAud = userValue; }

FunctVariable s_dPqPAud {
335 units = "none"; IOstatus = "input";
    getFunction = "get_sAud"; setFunction = "set_sAud";
}
real get_sAud() { return s_dPqPAud; }
void set_sAud(real userValue) { s_dPqPAud = userValue; }
340

//-----
// ***** OPTION VARIABLE SETUP *****
//-----

345 Option switchAud {
    allowedValues = { "BASE", "AUDIT" }

```

```

    description = "Determines if the audit factors are used";
    IOstatus = "input";
    trigger=TRUE;
350 }

Option switchBurn {
    allowedValues = { "FAR", "EQRATIO" }; // "FUEL", "WFUEL", "...
        TEMPERATURE", __ mod 18 Dec 2006 - IA - added "FILLFRACTION...
        "
355     description = "Switch determines if burner is running to fuel ...
        flow, FAR, or T4. Setting option to FUEL will burn using ...
        the burner value as an input. Setting the option to WFUEL ...
        will burn using the value coming in from the fuel station."...
        ;
    trigger=TRUE;
}

Option switchDes {
360     allowedValues = { "DESIGN", "OFFDESIGN" };
    description = "Design switch";
    trigger=FALSE;
}
// input kept in for backward compatible (remove later)
365 Option switchHotLoss {
    allowedValues = { "INPUT", "CALCULATE","input" };
    description = "Switch determines if the hot pressure loss is ...
        input or iterated on";
    trigger=TRUE;
}
370
//-----
// **** SETUP PORTS, FLOW STATIONS, SOCKETS, TABLES ****
//-----

375 // FLUID PORTS
FluidInputPort Fl_I {
    description = "Incoming flow";
}

380 FluidOutputPort Fl_01 {
    description = "Exiting combustion flow";
}

FluidOutputPort Fl_02 {
385     description = "Exiting bypass flow";
}

// FUEL PORTS

390 FuelInputPort Fu_I {
    description = "Incoming fuel flow";
}

```

```

}

// BLEED PORTS
395 // THERMAL PORTS

// MECHANICAL PORTS

400 // FLOW STATIONS

//_____flow stations modified 18 Dec 2006- IA
FlowStation Fl_Icomb {
405     description = "Inlet station to detonation tube section
of burner (after the initial pressure loss is applied)";
}

FlowStation Fl_IcombAir {
410     description = "Copy of the inlet station to detonation ...
tube
section of burner(after the initial pressure loss is ...
applied,
before flow is split and partitioned)";
}

FlowStation Fl_Iprg {
415     description = "Station containing detonation tube purge ...
fluid";
}

FlowStation Fl_Ocomb {
420     description = "Exit station to combustion section of burner
(before thermal storage heat transfer is calculated)";
}

FlowStation Fl_Vit {
425     description = "Vitiated Fluid flow station before detonation (...
cold)";
}
// _____-----end flow station modifications

// SOCKETS

430 Socket S_dPqP {
allowedValues = { "dPqPBase" };
description = "Dry duct and valve pressure loss"; //__ mod -...
IA- 18 Dec 2006
socketType = "dPqP";
}

435 Socket S_eff {
allowedValues = { "effBase", "effChemBase" };
description = "PulseDetonationCombustor adiabatic efficiency";
}

```

```

    socketType = "BURN_EFFICIENCY";
440 }

Socket S_Qhx {
    allowedValues = { "Qhx" };
    description = "Thermal storage socket";
445 socketType = "HEATTRANSFER";
}

// TABLES
450 //-----
// ***** INTERNAL SOLVER SETUP *****
//-----

455 //-----
// ***** ADD SOLVER INDEPENDENTS & DEPENDENTS *****
//-----

460 //-----
// ***** VARIABLE CHANGED METHODOLOGY *****
//-----
void variableChanged( string name, any oldVal ) {
    // Check to see what variables were changed....
    // Change input/output status as necessary - IA- 18 Dec 06
465
    if( name == "switchBurn" ) {
        if ( switchBurn == "FAR" ) {
            FAR.IOstatus = "input";
            Wfuel.IOstatus = "output";
470 TtCombOut.IOstatus = "output";
            eqRatio.IOstatus = "output";
        }
        //     else if ( switchBurn == "FUEL" ) {
        // FAR.IOstatus = "output";
475 // Wfuel.IOstatus = "input";
        // TtCombOut.IOstatus = "output";
        //     }
        //     else if ( switchBurn == "WFUEL" ) {
        // FAR.IOstatus = "output";
480 // Wfuel.IOstatus = "output";
        // TtCombOut.IOstatus = "output";
        //     }
        //----- added 5 Feb 2007 -IA-
        else if ( switchBurn == "EQRATIO" ) {
485 FAR.IOstatus = "output";
            Wfuel.IOstatus = "output";
            TtCombOut.IOstatus = "output";
            eqRatio.IOstatus = "input";
        }
490 //----- end of additions -IA-

```

```

}

else if( name == "switchHotLoss" ) {
  if ( switchHotLoss == "INPUT" ) {
495  dPqPRayleigh.IOstatus = "input";
    }
    else if ( switchHotLoss == "input" ){ switchHotLoss = "...
      INPUT"; }
    else {
500  dPqPRayleigh.IOstatus = "output";
    }
  }

  else if( name == "switchAud" ) {
505  a_dPqPAud.IOstatus = "inactive";
    s_dPqPAud.IOstatus = "inactive";
    if( switchAud == "AUDIT" ) {
      a_dPqPAud.IOstatus = "input";
      s_dPqPAud.IOstatus = "input";
510  }
    }
  }

//-----
515 // ***** PERFORM ENGINEERING CALCULATIONS *****
//-----

void calcPreLoss() {

520  //-----
  // Check to see if the pressure sockets are empty, if not ...
  // thenexecute
  //-----
  if ( !S_dPqP.isEmpty() ) {
525  S_dPqP.execute();
  }
  dPqP = dPqPBase * s_dPqP + a_dPqP; // calculate pressure ...
  losses (dry duct and Valve)
  if( switchDes == "OFFDESIGN" ) {
    if( switchAud == "AUDIT" ) {
530  dPqP = dPqP * s_dPqPAud + a_dPqPAud;
    }
  }

  //comment -IA- Collect total enthalpy at inlet
  real hin = Fl_I.ht;
535  real Pin = ( 1 - dPqP ) * Fl_I.Pt; //coment -IA- apply ...
  pressure losses as calculated above
  //comment -IA- copy flow to combustor flow
  Fl_Icomb.copyFlowStatic( "Fl_I" );
  Fl_Icomb.setTotal_hP( hin, Pin );

```

```

540 }

void calcBurn() {

    real TtCombOutTemp;
545 real htStoich;
    real WFuelLimit;
    real WFuelHeat;

    Fl_0comb.copyFlow( "Fl_Icomb" );

550 //-----
    // Efficiency
    //-----
    if ( !S_eff.isEmpty() ) {
555     S_eff.execute();
    }
    eff = effBase * s_eff + a_eff;
    effChem = effChemBase * s_effChem + a_effChem;

560 //-----
    // Burn
    //-----
    Fl_0comb.burn( "Fu_I", eff );

565 //-----
    // if inputting a PW type of efficiency adjust the temperature
    //-----
    if ( effChem < 1.0 ) {
570     TtCombOutTemp = effChem * ( Fl_0comb.Tt - Fl_Icomb.Tt ) +
        Fl_Icomb.Tt;
        Fl_0comb.setTotalTP( TtCombOutTemp, Fl_Icomb.Pt ); // use ...
        Pin
    }

}

575 void calcRayleighLoss() {

    flagRayleighChoked = 0;
    flagRayleighLossTooMuch = 0;

580 PqPRayleigh = 1.0;
    PqPRayleighError = 0.0;

585 //-----
    // self-convergent iteration loop for internal momentum ...
    // pressure drop calc
    //-----
    for( countRayleigh=0; countRayleigh<=countRayleighMax;

```

```

countRayleigh++) {
590
    //-----
    // input or output dPqPRayleigh
    //-----
595    if( switchHotLoss == "INPUT" ) {
PqPRayleigh = 1.0 - dPqPRayleigh;
    }
    else if( switchHotLoss == "CALCULATE" ) {
600    dPqPRayleigh = 1.0 - PqPRayleigh;
    }

    //-----
    // calculate momentum pressure drop
    //-----
605    real PtCombOut = PqPRayleigh * Fl_Icomb.Pt;

    Fl_Ocomb.setTotal_hP( Fl_Ocomb.ht, PtCombOut );

610
    //-----
    // Check momentum pressure drop
    //-----
    PqPRayleighNew = PqPRayleigh;
615
    if ( switchHotLoss == "CALCULATE" ) {

620
    //-----
    // make this thing a constant area burner
    //-----
    Fl_Ocomb.A = Fl_Icomb.A;
    flagRayleighChoked = 0;
    if( Fl_Ocomb.MN > 1.0 ) {
625
        // when MN > 1.0 FlowStation static calc is
        // not consistent with Area
        // Fl_Ocomb.MN = 1.0;
        // do not do this - creates major iteration problems
        flagRayleighChoked = 1;
630    }

    //-----
    // Calculate the exit static pressure from the momentum eqn
    // assume the fuel has the same velocity as the entrance flow
    //-----
635
    real PsMomMeth1;
    PsMomMeth1 = Fl_Icomb.W*Fl_Icomb.V - Fl_Ocomb.W*Fl_Ocomb.V;
    PsMomMeth1 = PsMomMeth1/C_GRAVITY;
640
    PsMomMeth1 = PsMomMeth1 + Fl_Icomb.Ps * Fl_Icomb.A;

```

```

PsMomMeth1 = PsMomMeth1/Fl_Ocomb.A;
real PsMomMeth2;
//PsMomMeth2 = Fl_Ocomb.W*Fl_Icomb.V;
PsMomMeth2 = Fl_Icomb.W*Fl_Icomb.V + Wfuel*Fl_Icomb.V*...
    fuelFractV;
645 PsMomMeth2 = PsMomMeth2/C_GRAVITY;
PsMomMeth2 = PsMomMeth2 + Fl_Icomb.Ps * Fl_Icomb.A;
PsMomMeth2 = PsMomMeth2/Fl_Ocomb.A;
PsMomMeth2 =
    PsMomMeth2/(1.0+Fl_Ocomb.gams*Fl_Ocomb.MN*Fl_Ocomb.MN);
650 //PsMomMeth1 = PsMonMeth2;

//-----
// Note Meth1 = Meth2 when MN <= 1.0
// Use Meth2 - seems more stable the Meth1 when MN > 1.0
//-----
655 PqPRayleighNew = (PsMomMeth2/Fl_Ocomb.Ps) * PqPRayleigh;
    }

    // Check against tolerance
660 PqPRayleighError = PqPRayleighNew - PqPRayleigh;
    if( abs(PqPRayleighError) < tolRayleigh ) { break; }

    // Bounding of PqPRayleigh movement to PqPRayleighStep
    real sign;
665 sign = PqPRayleighError/abs(PqPRayleighError);
    PqPRayleighDelta = sign *
min(abs(PqPRayleighError),PqPRayleighStep);
    PqPRayleighNew = PqPRayleigh + PqPRayleighDelta;

670 // Lower limit of PqPRayleigh - limit too much loss to ...
    PqPRayleighMin
    if( PqPRayleighNew < PqPRayleighMin ) {
if( flagRayleighLossTooMuch == 1 ) {
    ESOreport( 1023901,"Rayleigh pressure loss limited, too much...
        loss", FALSE );
    break;
675 }
PqPRayleighNew = PqPRayleighMin;
flagRayleighLossTooMuch = 1;
    }
    else {
680 flagRayleighLossTooMuch = 0;
    }

    /*
    // debug info
685 cout << Fl_Ocomb.A << " ";
    cout << Fl_Ocomb.MN << " ";
    cout << Fl_Ocomb.Ps << " ";
    cout << PsMomMeth1 << " ";
    cout << PsMomMeth2 << " ";

```

```

690     cout << PqPRayleigh << " ";
        cout << PqPRayleighNew << " ";
        cout << endl;
        */

695     //-----
        // check for convergence
        //-----
        if( countRayleigh >= countRayleighMax ) {
700     ES0report( 1023901,"Rayleigh iteration failed to converge, ...
            counter exceed max", FALSE );
        break;
        }

        PqPRayleigh = PqPRayleighNew;

705     }

        if( flagRayleighChoked == 1 ) {
710     ES0report( 1023901,"Rayleigh Fl_Icomb.MN exceed choked
            condition", FALSE );
        }
    }

715 void calculate() {

        //-----
        // Preburning pressure loss
        //-----
720     calcPreLoss(); // creates Fl_Icomb, applies pre-losses

        real FARin = Fl_Icomb.FAR;
        real WARin = Fl_Icomb.WAR;

725     if (Fl_I.MN == 0. && Fl_I.Aphy == 0.){
            Fl_Icomb.MN = 0.4;
            Fl_Icomb.setTotal_hP(Fl_Icomb.ht, Fl_Icomb.Pt);
        }

730     //-----
        // Pre-calculate Burning to obtain enthalpy,etc.
        //-----
        if ( switchBurn == "FAR" ) {
735     //-----
            // determine the fuel weight flow from the input FAR
            //-----
            Wfuel = ( Fl_Icomb.W/(1. + FARin +WARin))*(FAR -FARin);
            Fu_I.Wfuel = Wfuel;
740     eqRatio = FAR/Fu_I.FARst; // Added 5 Feb 2007 - IA

```

```

    calcBurn();
    calcRayleighLoss();

745     TtCombOut = Fl_0comb.Tt;

    }
    //#####
    // Added 5 February 2007 - IA
750    // do an equivalence ratio calculation
    else if (switchBurn == "EQRATIO") {
        FAR = eqRatio*Fu_I.FARst;
        Wfuel = (Fl_Icomb.W/(1.+ FARin+ WARin))*(FAR - FARin);
        Fu_I.Wfuel = Wfuel;

755        calcBurn();
        calcRayleighLoss();

        TtCombOut = Fl_0comb.Tt;

760    }

    //-----
    //make a flow station that has props of cold vitiated air
    //-----
765    Fl_Vit.copyFlowStatic("Fl_0comb");
    Fl_Vit.setTotalTP(Fl_Icomb.Tt, Fl_Icomb.Pt);

    //-----
    //copy inlet flow for pure air reference to be used later
770    //-----
    //Take a snapshot of air after it has entered detn tubes
    Fl_IcombAir.copyFlowStatic("Fl_Icomb");

    // Copy input flow properties for internal bypass flow
775    // - W set later
    Fl_02.copyFlow("Fl_IcombAir");

    //-----
    // On-design loop
780    //-----
    if (switchDes == "DESIGN"){

        //-----
        // Initialize local variables
785    //-----
        real uCJ, a_1, rhoVit, freq, PcqPi, errors;
        real gamt, Cpt, beta, MCJ2, PcqPi2;
        real Atube, Vtube;//, mCycle, Wtube;
        real MFP, Wvalve, gma_I;
790    real mFillAir, mPurgeAir, mPureAir;
        real tDetonation, tDetProp, tBlowdown, tPurge, tFill, iVel;
        real gam_s, gmm_fc;

```

```

real WtotAir, Wbypass;

795  int count;

//---- initiated but not iterated -----
//static density of cool vitiated fluid
rhoVit = Fl_Vit.rhot;  //(lbm/ft^3)
800

// speed of sound in pure air, stagnated in detonation
// tube that the detonation wave propogates in to
a_1 = sqrt(Fl_Icomb.gamt*Fl_Icomb.Rt*Fl_Icomb.Tt*25037.);

805

// =====
// Calculate Chapman-Jouguet Mach number for wave
// as described in Heiser and Pratt
//=====
810 //*** input variables: //
//*** output variables: //MCJ, deltaS, qadd //
//*** Flow Stations: //Fl_0comb, Fl_Icomb //
// local variables: //gamt, Cpt, qadd, beta, MCJ2 //

815 //----- Arithmetically average specific heats -----
// arithmetic mean of gamma for stopped fluid
gamt = (Fl_0comb.gamt + Fl_Icomb.gamt)/2.0;

// arithmetic mean of Cp for a stopped fluid
820 Cpt = (Fl_0comb.Cpt + Fl_Icomb.Cpt)/2.0;

//----- Calculate heat addition per Heiser-Pratt cycle ---
// calculate non-dimensional heat addition
825 qadd = (Fl_0comb.ht - Fl_Icomb.ht)/(Cpt*Fl_Icomb.Tt);

//----- Calculate Chapman-Jouget Mach number -----
beta = (gamt + 1.0)*qadd+1.0;
MCJ2 = beta + sqrt( beta**2 - 1.0 );
MCJ = sqrt(MCJ2);
830

//----- Calculate Entropy gain based on CJ detonation -
deltaS = Cpt*(-log(MCJ2*((gamt+1.0)/
(1.0+gamt*MCJ2))*((gamt+1.0)/gamt)) );

835 //---- calculate the pressure rise using the H & P method --
PcqPi = (1.0+ gamt*MCJ2)/(gamt+1.0);
uCJ = a_1*MCJ;

840 //----- Calculate tube volume and Area -----
Atube = (PI/4.)*dTube**2/144.; // ft^2
Vtube = Atube*(lTube/12); // ft^3

//----- calculate the valve inlet mass flow rate -----

```

```

845   gma_I=Fl_IcombAir.gamt;
      MFP = Mvalve*sqrt( (gma_I*32.174)/(Fl_IcombAir.Rt*778.16) )
          *(1.+(gma_I-1.)/2.*Mvalve**2)**( (gma_I+1.)/(2.*(1.-gma_I)...
          ));
      Wvalve = (Fl_IcombAir.Pt/sqrt(Fl_IcombAir.Tt))
          *(Atube*144.*ARvalve)*MFP;
850
      //-----
      // On-Design: Calculate bypass ratio
      //-----
      //*** input Variables: //dTube, lTube, n_tubes, fillFrac
855   //          // purgeFrac,
      //*** iterated Variables // freq
      //*** output Variables: // iBPR
      //*** local variables: //WfillAir, WpurgeAir, WpureAir, ...
          WtotAir
      //          Wbypass, WpurgeAir, Wvit, //
860   //*** Flow Stations:    // Fl_IcombAir, Fl_Icomb, Fl_Iprg, ...
          Fl_Vit, //

      //---- Calculate the split and partition of flow -----
      // amount of air that will be mixed with fuel - 1 tube
      mFillAir = Vtube*(rhoVit*fillFrac)/(1.+FAR);
865
      // amount of air that will purge during each cycle - 1 tube
      mPurgeAir = Vtube*(Fl_IcombAir.rhos*purgeFrac);

      // total air per cycle flowing though one tube
870   mPureAir = mFillAir + mPurgeAir;

      //-----
      // Timing - calculate frequency
      //-----
875   //*** input Variables: // DDT, tValve, Ltube, ff, pf, tCycle
      //*** iterated Variables: // uCJ, PcqPi
      //*** output Variables // tCycle, tauValveOpen, freq
      //*** local variables: // tDetonation, tDetProp, tBlowdown, ...
          tPurge,
      //          // tFill
880   //-----

      //----- Detonation time -----
      // DetProp time is relatively independant of fill fractn
885   tDetProp= lTube/(uCJ*12);

      //DDT is input, tDetonationPropogatio calcd
      // (may need to iterate)
      tDetonation = DDT + tDetProp;
890
      //----- Blowdown time -----
      // assume choked flow at tube exit and calculateon

```

```

// blowdown based draw-down time of a pressurized
// tank calculated on pressure differential
895 gam_s = Fl_IcombAir.gams; // larger gamma is more conservative
gmm_fc = ((gam_s + 1.)/2.)*(-(gam_s+1.)/( 2.*(gam_s-1.)) ); ...
//

//#### tBlowdown: Use ~1/2 calcd pressure (to match experimental ...
data)
// we'll use CJ det wave velocity as the speed of sound in the...
gas
900 // since a cannot be directly calc'd
// note tBlowdown is proportional to tube length
// tauBlDn is proportional to tube length...
tBlowdown = (log(0.4*PcqPi)/gmm_fc)*(lTube/uCJ);

905 //----- Fill and Purge time -----
// Use the choked flow at valve inlet and the mass flow rate ...
as
// calculated outside the loop to calculate fill time (m/ mdot...
)
tPurge = tValve + mPurgeAir/Wvalve; //(s)
tFill = tValve + mFillAir/Wvalve; //(s)
910 //Improvement could be made by calculating vitiated air ...
velocity...

//----- Cycle Time output calculation -----
tCycle = tDetonation + tBlowdown + tPurge + tFill;
tauValveOpen = (tPurge+tFill)/tCycle;
915 freq = 1./tCycle;
//cout << "\n \n tDetonation, tBlowdown, tPurge, tFill PcqPi...
"<<" "<< tDetonation <<" "<< tBlowdown<<" "<< tPurge<<" "<< ...
tFill<<" freq" << 1/tCycle << " " << PcqPi << endl;

//----- Set total mass flow through tubes -----
WtotAir = mPureAir*n_tubes*freq;
920 // steady-state flow rate into tubes

// conservation of mass check
if (WtotAir > Fl_I.W) {

925 fillFrac = fillFrac*(Fl_I.W/WtotAir);
purgeFrac = purgeFrac*(Fl_I.W/WtotAir);

mFillAir = Vtube*(rhoVit*fillFrac)/(1.+FAR);
// amount of air that will be mixed with fuel - 1 tube
930 mPurgeAir = Vtube*(Fl_IcombAir.rhos*purgeFrac);
// amount of air that will purge during each cycle -1 tube

mPureAir = mFillAir + mPurgeAir;
935 // total air per cycle flowing though one tube

```

```

WtotAir = Fl_I.W;
cout << " !pf & ff changed to: " << purgeFrac << " " << ...
    fillFrac << endl;
}
940 //----- Set iBPR -----
    Wbypass = Fl_I.W - WtotAir;
    // steady-state flow rate sent to bypass

945    iBPR = Wbypass/WtotAir;
    // steady-state internal PDC bypass ratio

    iBPRdes = iBPR;

950 //----- Set bypass exit flow SPLIT -----
    Fl_O2.W = Wbypass;

    //----- Set purge and fill stations PARTITION -----
    Fl_Iprg.copyFlowStatic("Fl_IcombAir");
955 // copy flow for purge function

    // ----- PURGE AIR -----
    Fl_Iprg.AphyDes = (Atube*144)*n_tubes;//Set phys area
    Fl_Iprg.W = mPurgeAir*freq*n_tubes; //set m dot

960 // ----- FILL AIR -----
    Fl_Icomb.copyFlow("Fl_IcombAir");
    Fl_Icomb.AphyDes = Atube*144.*n_tubes*tauValveOpen;
    // Actual area is multiplied by tauVO to get equivalent
965 // area. - Fluid flows steadily through this area

    Fl_Icomb.W = mFillAir*n_tubes*freq; //
    Fl_Icomb.setTotal_hP(Fl_IcombAir.ht, Fl_IcombAir.Pt);
    //sets time-averaged static conditions

970 //-----
    // Burning
    //-----
    // FAR was calculated prior to enteringh this
975 // point - so we just need to modify
    // Wfuel based on changed Fl_Icomb.W
    Wfuel = ( Fl_Icomb.W / ( 1. + FARin + WARin))*( FAR - FARin...
        );
    Fu_I.Wfuel = Wfuel;

980    calcBurn();
    calcRayleighLoss();

    TtCombOut = Fl_Ocomb.Tt;

985 //=====
    // Apply Dyer-Kaemming correction to obtain tube flow

```

```

// at exit(ignores the kinetic energy of shock wave.)
//=====
990 Fl_0comb.setTotal_hS(Fl_0comb.ht, Fl_Icomb.S+deltaS);
}

// OFF-DESIGN CODE GOES HERE

995 //-----
// Add split flows back to combusted flow
//-----
Fl_0comb.add("Fl_Iprg"); //add purge flow in (uncorrected)

1000 //=====
// Apply corrections to the flow for transition to ...
// steady state (TTSS)
//=====
1005 //*** local Variables: // Snew, Pnew
//*** Input Variables: // deltaS, TTSSeff, TTSSdPqP
//*** Flwo stations: // Fl_0comb, Fl_Vit
real hnew, Pnew; //

//----- Calculate new Entropy and Pressure -----
1010 //
// eff = (dht)TTSF/(dht)comb + 1.
// current h - ( h gained)*(1.-eff)
hnew = Fl_0comb.ht - (Fl_0comb.ht -
Fl_Icomb.ht)*(1.0-TTSSeff);
Pnew = Fl_0comb.Pt*(1.0-TTSSdPqP);

1015 //End of 12Jan2007 additinos - IA
//#####
Fl_01.copyFlow( "Fl_0comb" );

1020 //----- update fluid properties based on new S & P
Fl_01.setTotal_hP(hnew, Pnew); //added 12Jan2007 - IA

//-----
1025 // Thermal storage calculations
//-----
if ( !S_Qhx.isEmpty() ) {
S_Qhx.execute();
}

1030 real hout = Fl_01.ht - Qhx / Fl_01.W;
Fl_01.setTotal_hP( hout, Fl_01.Pt );

//-----
1035 // store the design value of FAR for use in guessing
//-----
if ( switchDes == "DESIGN" ) {
FARDes = FAR;
}

```

```
    }
1040 }

//-----
// register the appropriate errors at build time
1045 //-----
void VCinit()
{
    ES0regCreate( 1023901, 8, "", TRUE, FALSE, TRUE ); // ...
        provisional
    ES0regCreate( 1093901, 8, "", TRUE, FALSE, TRUE ); // ...
        provisional
1050 }
}

#endif
```

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Vita

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He studied at Idaho State University and Embry-Riddle Aeronautical University before transferring to Utah State University where he completed the rigorous Mechanical Engineering curriculum, receiving his Bachelor of Science degree in December of 2002. While earning his undergraduate degree he worked as a student researcher for the Space Dynamics Laboratory in the science group. As part of his work duties he was able to co-author several conference papers submitted to SPIE. He also authored a paper on orbital dynamics that was selected by AIAA to be presented at the 2002 World Space Congress student paper competition.

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